

Time-Temperature-Precipitation Behavior in Al-Li Alloy 2195

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TABLE OF CONTENTS

1. INTRODUCTION	1
2. TECHNICAL APPROACH	2
3. EXPERIMENTAL PROCEDURES	3
3.1 Processing	3
4. RESULTS AND DISCUSSION	4
4.1 GP Zone, θ'' (Al ₂ Cu), and δ' (Al ₃ Zr) 4.2 T_1 (Al ₂ CuLi) 4.3 θ' (Al ₂ Cu) and θ (Al ₂ Cu) 4.4 T_2 (Al ₅ Li ₃ Cu) and T_B (Al _{7.5} LiCu ₄) 4.5 Time-Temperature-Precipitation Diagrams	4 5 6 7
5. SUMMARY	11
REFERENCES	13

LIST OF FIGURES

1.	Heat-treatment matrix for solution-treated Al-Li 2195		
2.	Hardness variations as a function of heat-treatment conditions	4	
3.	Matrix consisting of entangled dislocations and dispersed δ' particles after aging at 200 °F/1 hr (dissolved GP zone and θ'' due to reversion)	5	
4.	T_1 nucleation occurring preferentially at subgrain boundaries at 300 °F	5	
5.	(a) Subgrain boundary T_1 , which grew longer and thicker than matrix T_1 , and (b) significant occurrence of nucleation and growth of T_1	6	
6.	Microstructure primarily consisting of matrix θ' and grain boundary θ after heat treatment at 700 °F/1 hr	6	
7.	θ' nucleation diminishing significantly at 800 °F	7	
8.	T_{2} and $T_{\rm B}$ forming at 800 °F/0.1 hr	7	
9.	TTP diagrams for Al-Li 2195	8	
10.	Enhanced θ'' and T_1 nucleation in the matrix (rather than at subgrain boundaries) during the first step of TS aging (270 °F/20 hr)	9	
11.	(a) No T_1 precipitation seen at subgrain boundaries after TS aging as compared to (b) preferential T_1 precipitation seen at subgrain boundaries after conventional aging	10	
12.	Significantly enhanced CFT of Al-Li 2195 after TS aging	10	

LIST OF ACRONYMS AND SYMBOLS

δ' Al₃Li precipitate compound

 θ , θ' , θ'' Al₂Cu precipitate compounds

Ag silver

Al aluminum

CFT cryogenic fracture toughness

Cu copper

GP zone Guinier-Preston zone

HAZ heat-affected zone

HR_B Rockwell hardness B scale

Li lithium

Mg magnesium

SLWT super lightweight tank

T₁ Al₂CuLi strengthening precipitate

T₂ Al₅Li₃Cu strengthening precipitate

 $T_{\rm B}$ Al_{7.5}LiCu₄ strengthening precipitate

TEM transmission electron microscopy

TS two step

TTP time-temperature-precipitation

Zr zirconium

TECHNICAL MEMORANDUM

TIME-TEMPERATURE-PRECIPITATION BEHAVIOR IN AL-LI ALLOY 2195

1. INTRODUCTION

Transmission electron microscopy (TEM) was used to study time-temperature-precipitation (TTP) behavior in aluminum-lithium (Al-Li) 2195 alloy. Al-Li 2195 (nominally Al + 4 percent Cu + 1 percent Li + 0.3 percent Ag + 0.3 percent Mg + 0.1 percent Zr) was initially solutionized for 1 hr at 950 °F and then stretched 3 percent. Heat treatments were conducted for up to 100 hr at temperatures ranging from 200 to 1,000 °F. TTP diagrams were determined for both matrix and subgrain boundaries. Depending upon heat treatment conditions, precipitate phases (such as Guinier-Preston (GP) zone, θ'' , θ' , θ , δ' , T_1 , T_B , and T_2) were found in the alloy. The TTP diagrams were applied as a guide to avoid T_1 precipitation at subgrain boundaries, as part of an effort to improve the alloy's cryogenic fracture toughness (CFT). New understanding of TTP behavior was instrumental in the development of a two-step (TS) artificial aging treatment that significantly enhances CFT in Al-Li 2195.

2. TECHNICAL APPROACH

Due to its low density, high modulus, and good cryogenic properties, Al-Li 2195 was selected as the main structural alloy for the Space Shuttle's external super lightweight tank (SLWT). Al-Li 2195 has significantly higher strength than conventional 2xxx alloys (such as Al 2219) at both ambient and cryogenic temperatures, and it can be strengthened further through the use of an aging treatment that precipitates the primary strengthening precipitate T_1 (Al₂CuLi). Other phases (such as GP zone, θ'' , θ' , θ , and δ') are present after the alloy is artificially aged. Some work has been done on the precipitation characteristics of T_1 in order to optimize materials strength and fracture toughness. However, no systematic study on TTP behavior had previously been undertaken.

TTP behavior was studied at temperatures <400 °F, in part because a recent study 1,2 had indicated that CFT is related to the density, size, and location of T_1 . CFT decreases considerably as T_1 increases in density at the subgrain boundaries, but such precipitation could perhaps be minimized by an advanced aging treatment based upon a better understanding of TTP behavior in Al-Li 2195. TTP was also studied at higher temperatures, due to the importance of alloy properties in the heat-affected zone (HAZ), which is subjected to thermal cycles during weld repair. Since any additional thermal cycles will affect precipitate stability and transformation of the structural alloy, TTP diagrams must be available for comparison to welded microstructures in order to understand the thermal history of each SLWT.

This study was conducted to develop TTP diagrams for solution-treated and stretched Al-Li 2195, which will allow its precipitation reactions to be predicted at any aging temperature. Such diagrams will serve as a guide to optimize the alloy's mechanical properties by varying heat treatment parameters. This understanding will assist in the development of welding parameters to further improve the mechanical properties of Al-Li 2195 for use in SLWT applications.

3. EXPERIMENTAL PROCEDURES

3.1 Processing

The alloy (nominally Al-4.19Cu-0.95Li-0.29Mg-0.31Ag-0.12Zr) used in this study was processed by Reynolds and supplied in the form of 1.7-in-thick rolled plates. Rolling was conducted on an initial ingot thickness of ≈ 13 in, in the temperature range of 700 to 800 °F. The plates were then given a solutionizing treatment for 1 hr at 950 °F and then stretched 3 percent at ambient temperature. After stretching, the specimens were heat treated in the time-temperature envelope of 0.1 to 100 hr at 200 to 1,000 °F (see fig. 1). After heat treatment, hardness was measured for each specimen using the Rockwell hardness B scale (HR_B).

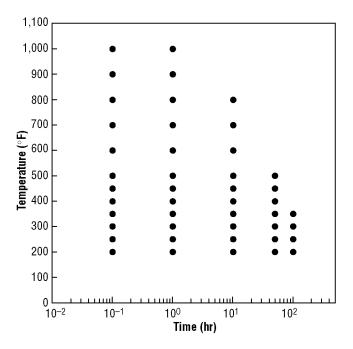


Figure 1. Heat-treatment matrix for solution-treated Al-Li 2195.

3.2 Microstructural Characterization

Specimens were sliced from the plates and mechanically ground to the desired thickness for microstructural examination, which was carried out using a JEOL, Ltd. 2000F transmission electron microscope operated at 200 kV. Samples were twinjet electropolished to perforation at –20 °F and 12 V in an electrolyte of 30-percent nitric acid in methanol. Precipitates were examined by the combined use of selected area diffraction and bright field/dark field techniques. Matrix and subgrain boundary precipitates were examined using an electron beam direction near the [110]_{matrix} zone axis.

4. RESULTS AND DISCUSSION

The Al-Li 2195 plate had a predominantly unrecrystallized microstructure, with coarse and pancake-shaped grains elongated along the rolling direction. The as-solution-treated and stretched microstructure primarily contained *GP* zone (localized concentrations of Cu atoms) due to natural aging,⁵ but was free of other secondary phases. Depending on heat-treatment conditions, the solution-treated material can decompose and precipitate via several different reactions. Therefore, hardness can vary as a function of heat-treatment conditions (see fig. 2).

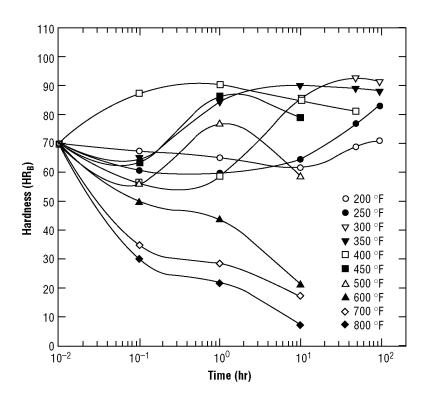


Figure 2. Hardness variations as a function of heat-treatment conditions.

4.1 GP Zone, θ'' (Al₂Cu), and δ' (Al₃Zr)

GP zone, θ'' , and δ' were the primary precipitate phases at 200 and 250 °F. During early stages of heat treatment (<500 °F), reversion caused a sharp decrease in hardness⁵ associated with some dissolution of GP zones and δ' . At 200 and 250 °F, the matrix microstructure consisted of primary entangled dislocations and dispersed δ' particles. No sign of T_1 nucleation was found under these conditions (see fig. 3). After longer heat treatment at 250 °F, a slight increase in hardness was noted, due to continual growth of θ'' .

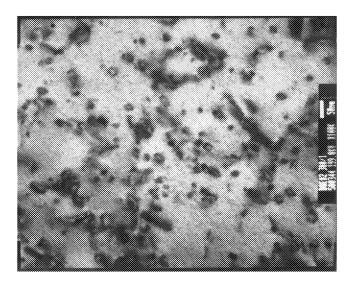


Figure 3. Matrix consisting of entangled dislocations and dispersed δ' particles after aging at 200 °F/1 hr (dissolved *GP* zone and θ'' due to reversion).

4.2 *T*₁ (Al₂CuLi)

 T_1 was the predominant precipitate at temperatures ranging from 280 to 500 °F. With its plate-shaped morphology, T_1 is the primary phase responsible for the increased hardness at temperatures <500 °F. It has a very strong tendency to nucleate at subgrain boundaries and grow very rapidly there (see fig. 4) in the same temperature range as T_1 formed in the matrix. However, subgrain T_1 is usually thicker, shorter, and has a smaller diameter than matrix T_1 . Only when the habit plane of T_1 was nearly parallel to the subgrain boundary plane did the precipitate grow very long (see fig. 5). In the matrix, T_1 grew continuously at the expense of GP zones, θ' , and θ'' . Exposure for a short time (6 min) at 350 and 400 °F led to abundant nucleation and significant coarsening of T_1 (see fig. 6).

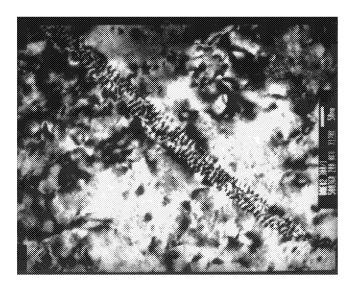


Figure 4. T_1 nucleation occurring preferentially at subgrain boundaries at 300 °F.

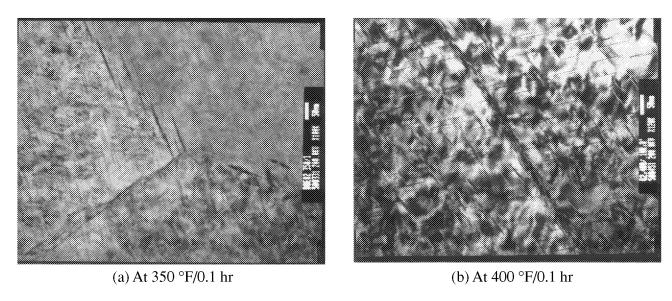


Figure 5. (a) Subgrain boundary T_1 , which grew longer and thicker than matrix T_1 , and (b) significant occurrence of nucleation and growth of T_1 .

4.3 θ' (Al₂Cu) and θ (Al₂Cu)

In the early stages of heat treatment at elevated temperatures (600 to 700 °F), θ' became the dominant precipitate (see fig. 6). Prolonged exposure at 700 °F led to the precipitation of equilibrium phases θ in the matrix and grain boundaries. Above 800 °F, the formation of incoherent θ accompanied the recovery and recrystallization processes, as evidenced by increased dislocation pileup and grain growth. At 800 °F, θ' nucleation diminished significantly (see fig. 7).



Figure 6. Microstructure primarily consisting of matrix θ' and grain boundary θ after heat treatment at 700 °F/1 hr.



Figure 7. θ' nucleation diminishing significantly at 800 °F.

4.4 T_2 (Al₅Li₃Cu) and T_B (Al_{7.5}LiCu₄)

 θ' , θ , T_2 , and T_B formed in the matrix and subgrain boundaries after 10 hr at 700 and 800 °F. Prolonged exposure at 800 °F led to the precipitation of equilibrium phases T_B (primarily in the matrix and grain boundaries) and T_2 (primarily at grain boundaries). T_B precipitated in the matrix and appeared to grow at the expense of θ' and θ , whereas T_2 led to continuous grain boundary coverage by coexisting with θ at grain boundaries (see fig. 8).

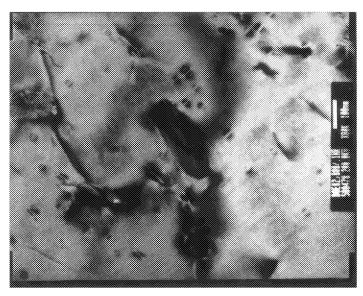


Figure 8. T_2 and $T_{\rm B}$ forming at 800 °F/0.1 hr.

At temperatures >600 °F, a significant drop in hardness occurred, which can be explained by insufficient nucleation of T_1 and precipitation of incoherent equilibrium phases θ , T_B , and T_2 . The number of subgrains also diminished due to extensive recovery and early stage of recrystallization. At temperatures >900 °F, more significant recrystallization and grain growth occurred, creating a microstructure essentially devoid of any strengthening precipitates in both the matrix and grain boundaries.

4.5 Time-Temperature-Precipitation Diagrams

The TTP diagrams consisted of C-curves for T_1 , θ' , θ'' , θ , T_2 , and T_B in the matrix and T_1 , θ' , θ , T_2 , and T_B at subgrain boundaries (see fig. 9). The phase boundaries (estimated by curve-fitting approach from TEM observation of the experimental samples) are given only for the start of each precipitation, and completion curves were not determined. Therefore, the region to the right of a curve represents the time-temperature envelope for which that phase was present in the microstructure after heat treatment.

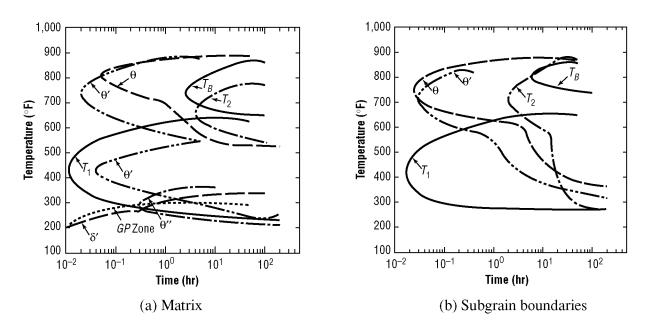


Figure 9. TTP diagrams for Al-Li 2195.

A very short but finite nucleation time (<6 min) exists for the onset of T_1 precipitation. In most cases, subgrain boundaries are favorable nucleation sites for T_1 at temperatures ranging from \approx 280 to 500 °F. Nucleation of T_1 is very rapid and, once nucleated, subgrain boundary T_1 always grows faster than matrix T_1 . The only ways found to avoid preferential T_1 precipitation at subgrain boundaries were to age the material <280 °F (the temperature at which T_1 forms) or enhance T_1 nucleation and growth in the matrix so that T_1 -forming elements (such as Li and Cu) became depleted at subgrain boundaries.

Caution should be exercised when using TTP diagrams for Al-Li 2195, since the energy state (e.g., grain size, hot working history, solution treatment temperature, degree of stretch) can shift relationships to the left or right. The commercial solution heat treatment commonly used for Al-Li 2195 (950 $^{\circ}$ F) would be expected to result in complete dissolution of all phases.

Although precipitation behavior may be affected by composition variations from heat to heat, these TTP diagrams can be used as a guide to select appropriate heat treatments for different service applications. If high strength and high fracture toughness are needed, then aging treatments should be selected from diagrams which precipitate maximum matrix T_1 for strength while avoiding or minimizing precipitation of subgrain boundary T_1 .

CFT and cryogenic strength are considered critical for the SLWT (which houses liquid oxygen and liquid hydrogen), as is the need for higher strength and fracture toughness at cryogenic temperatures than at ambient temperature (in order to avoid expensive cryogenic proof testing). Unfortunately, some commercial 2195 material was disqualified from the SLWT program when it exhibited unexpectedly low CFT which proved to be related to the density, size, and location of T_1 . Attempts were made to improve the fracture toughness of such material by reducing subgrain boundary T_1 while enhancing the nucleation of matrix T_1 , based on the observation that CFT decreases considerably as T_1 increases in density at the subgrain boundaries.

Since the TTP diagrams indicate that T_1 nucleation and growth can be controlled by avoiding temperature ranges that activate nucleation, a TS aging treatment was developed to enhance CFT.⁴ The initial aging sequence (20 hr at 270 °F) promotes θ'' and T_1 nucleation and growth in the matrix, while reducing T_1 nucleation at subgrain boundaries (see fig. 10). T_1 will eventually nucleate at subgrain boundaries and start to grow as aging continues (45 hr at 280 °F). However, the TS aging treatment allows T_1 to precipitate and grow in the matrix before it can develop in the subgrain boundaries. The early coarsening of matrix T_1 greatly reduces the concentration of Cu and Li in the matrix and thus hinders the growth of subgrain boundary T_1 in a diluted Al-Cu-Li solid solution. The higher temperatures used in conventional aging (32 hr at 290 °F) are not encountered during TS aging, which helps to constrain T_1 nucleation and growth at subgrain boundaries. As a result, a TS-aged specimen has a significantly different microstructure from that of a conventionally aged specimen (see fig. 11). Tests have indicated that TS aging can enhance CFT by as much as 30 percent (see fig. 12).

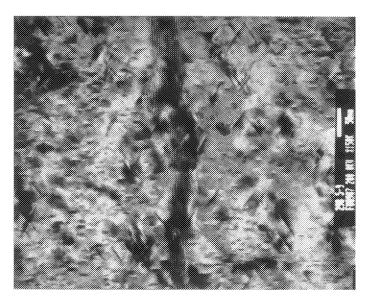


Figure 10. Enhanced θ'' and T_1 nucleation in the matrix (rather than at subgrain boundaries) during the first step of TS aging (270 °F/20 hr).

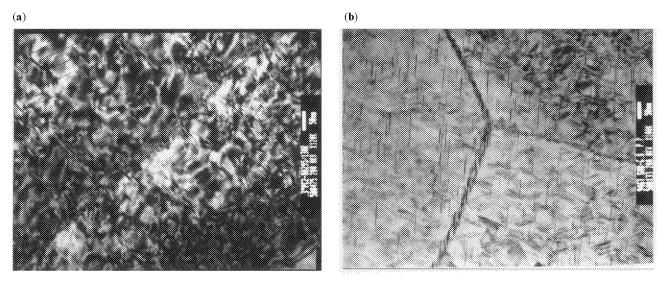


Figure 11. (a) No T_1 precipitation seen at subgrain boundaries after TS aging as compared to (b) preferential T_1 precipitation seen at subgrain boundaries after conventional aging.

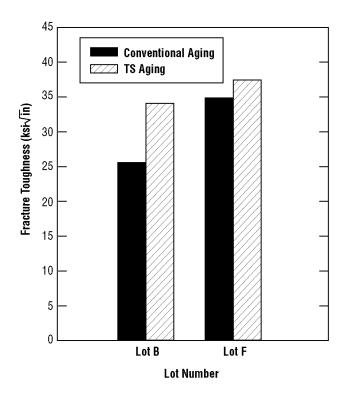


Figure 12. Significantly enhanced CFT of Al-Li 2195 after TS aging.

5. SUMMARY

- 1. TTP diagrams were determined for Al-Li 2195. They consisted of six separate C-curves for the matrix and four C-curves for the subgrain boundaries.
- 2. T_1 was the primary strengthening phase at temperatures <500 °F.
- 3. At temperatures <600 °F, T_1 was the only phase present at the subgrain boundaries. At temperatures between 300 and 500 °F, initial nucleation of T_1 occurred preferentially at the subgrain boundaries, with more T_1 growth seen there than in the matrix.
- 4. At temperatures >600 °F, θ' , θ , T_2 , and T_B formed at the subgrain boundaries and in the matrix. Prolonged exposure at 800 °F resulted in the precipitation of equilibrium phases T_B and T_2 . Apparently, T_B grew at the expense of θ' and θ , whereas T_2 coexisted with θ at the grain boundaries (which led to continuous grain boundary coverage).
- 5. At temperatures >600 °F, a significant drop in hardness occurred, which can be explained by insufficient nucleation of T_1 and precipitation of incoherent equilibrium phases θ , T_B , and T_2 . At temperatures >900 °F, significant recrystallization and grain growth resulted in a microstructure that was essentially devoid of any strengthening precipitates.
- 6. This study developed TTP diagrams which were instrumental in the development of an advanced aging treatment for Al-Li 2195 that is now being used in the SLWT program. This aging treatment has proved to be capable of improving the size, distribution, and density of T_1 in Al-Li 2195 and hence improving the alloy's CFT by up to 30 percent.

REFERENCES

- 1. Chen, P.S.; and Stanton, W.P.: "A New Aging Treatment for Improving Cryogenic Toughness of the Main Structural Alloy of the Super Lightweight Tank," *NASA Technical Memorandum* 108524, 1996.
- 2. Chen, P.S.; Kuruvilla, A.K.; Malone, T.W.; and Stanton, W.P.: "The Effects of Artificial Aging on the Microstructure and Fracture Toughness of Al-Cu-Li Alloy 2195," *J. Mater. Eng. & Per.*, Vol. 7, p. 682, 1998.
- 3. Blankenship, Jr., C.P.; and Starke, Jr., E.A.: "Structure-Property Relationships in Al-Li-Cu-Mg-Ag-Zr Alloy X2095," *Acta. Metall.*, Vol. 42, p. 845, 1995.
- 4. Chen, P.S.; and Stanton, W.P.: "Artificial Aging Effects on Cryogenic Fracture Toughness of the Main Structural Alloy for the Super Lightweight Tank," NASA Technical Memorandum, publication pending, 2002.
- 5. Pickens, J.R.; Kumar, K.S.; Brown, S.A.; et al.: "Evaluation of the Microstructure of Al-Cu-Li-Ag-Mg WeldaliteTM Alloys," *NASA Contractor Report 4368*, 1991.

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