

# Comparative Measurements of Earth and Martian Entry Environments in the NASA Langley HYMETS Facility

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Arc-jet facilities play a major role in the development of heat shield materials for entry vehicles because they are capable of producing representative high-enthalpy flow environments. Arc-jet test data is used to certify material performance for a particular mission and to validate or calibrate models of material response during atmospheric entry. Materials used on missions entering Earth's atmosphere are certified in an arc-jet using a simulated air entry environment. Materials used on missions entering the Martian atmosphere should be certified in an arc-jet using a simulated Martian atmosphere entry environment, which requires the use of carbon dioxide. Carbon dioxide has not been used as a test gas in a United States arc-jet facility since the early 1970's during the certification of materials for the Viking Missions. Materials certified for the Viking missions have been used on every entry mission to Mars since that time. The use of carbon dioxide as a test gas in an arc-jet is again of interest to the thermal protection system community for certification of new heat shield materials that can increase the landed mass capability for Mars bound missions beyond that of Viking and Pathfinder. This paper describes the modification, operation, and performance of the Hypersonic Materials Environmental Test System (HYMETS) arc-jet facility with carbon dioxide as a test gas. A basic comparison of heat fluxes, various bulk properties, and performance characteristics for various Earth and Martian entry environments in HYMETS is provided. The Earth and Martian entry environments consist of a standard Earth atmosphere, an oxygen-rich Earth atmosphere, and a simulated Martian atmosphere. Finally, a preliminary comparison of the HYMETS arc-jet facility to several European plasma facilities is made to place the HYMETS facility in a more global context of arc-jet testing capability.

## Nomenclature

$C_1$	= Conversion Coefficient, 9.48E-4, Btu/(s.W)
$C_2$	= Conversion Coefficient, 2.23E-3, (min.ft <sup>3</sup> )/(s.gal)
$C_P$	= Specific Heat Capacity, Btu/(lb <sub>m</sub> .°R)
$CW$	= Cooling Water Flow, gal/min
$E$	= Voltage, Volts
$H_O$	= Bulk Enthalpy, Btu/lb <sub>m</sub>
$I$	= Current, Amps
$l$	= Length, ft
$m_T$	= Mass Loss Rate of Teflon <sup>®</sup> , lb <sub>m</sub> /(ft <sup>2</sup> .s)

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$M_{Total}$	=	Mass Flow Rate, lb <sub>m</sub> /s
$P_{stag}$	=	Stagnation Pressure, atm
$q_{FCCW}$	=	Fully-Catalytic Cold Wall Heat Flux Rate, Btu/(ft <sup>2</sup> .s)
$q_{NCCW}$	=	Non-Catalytic Cold Wall Heat Flux Rate, Btu/(ft <sup>2</sup> .s)
$R_{eff}$	=	Effective Hemispherical Radius, ft
$R_{Flat\ Face}$	=	Flat Face Radius, ft
$t$	=	time, s
$T$	=	Temperature, °R
$T_{CW}$	=	Cooling Water Temperature, °R
$T_{FCCW}$	=	Fully-Catalytic Cold Wall Temperature, °R
$\Delta$	=	Change
$\rho$	=	Density, lb <sub>m</sub> /ft <sup>3</sup>

## I. Introduction

Arc-jet facilities play a major role in the development of heat shield materials for entry vehicles because they are capable of producing representative high-enthalpy flow environments. Arc-jet test data is used to certify material performance for a particular mission and to validate or calibrate models of material response during atmospheric entry. Extensive arc-jet testing was conducted during the development of the Superlight Ablator, known as SLA-561V, which was used on the forebody heat shield of every entry mission to Mars: Viking Landers 1 and 2 in 1976, Mars Pathfinder in 1997, Mars Polar Lander in 1999, Mars Exploration Rovers' Spirit and Opportunity in 2004, and Mars Phoenix Lander in 2008.

The arc-jet test database generated in the late 1960's and early 1970's on SLA-561V for the Viking Landers included tests in a standard Earth atmosphere consisting of 79% Nitrogen (N<sub>2</sub>) and 21% Oxygen (O<sub>2</sub>) by volume (or equivalently moles) and two simulated Martian atmospheres: 1) 72% N<sub>2</sub> and 28% Carbon Dioxide (CO<sub>2</sub>) and 2) 50% N<sub>2</sub> and 50% CO<sub>2</sub> by volume.<sup>1-3</sup> The Viking database was expanded in the early 1990's to include the higher entry heating environment for the Mars Pathfinder mission.<sup>4,5</sup> Certification of SLA-561V for use on the Mars Pathfinder mission was performed using only a standard Earth atmosphere because CO<sub>2</sub> test capability no longer existed in the United States. All subsequent Mars missions relied on the expanded database from which ablation response models were calibrated.

The Mars Science Laboratory (MSL), scheduled for launch in late 2011, represents a significant increase in size and landed mass over that of previous missions. The resulting increase in the entry heating environment again required the certification of SLA-561V for use on the forebody heat shield. During arc-jet certification, SLA-561V suffered a catastrophic failure at conditions within the flight envelope for MSL. The MSL program then selected Phenolic Impregnated Carbon Ablator, or PICA, for use on the forebody heatshield<sup>6,7</sup> as it was successfully used for the Stardust return capsule and was one of two ablators being considered for the Orion Crew Exploration Vehicle (CEV). The CEV arc-jet database and response models for PICA were expanded to include the Martian atmospheric entry conditions predicted for MSL. To compensate for the fact that testing in CO<sub>2</sub> could not be performed, response models for PICA were indirectly validated by varying the percentage of oxygen<sup>7</sup> in arc-jet tests performed at the NASA Johnson Space Center (JSC). One of the differences between tests in air and CO<sub>2</sub> is the amount of available oxygen and it is unknown whether an oxygen-rich environment is comparable to a CO<sub>2</sub> environment.

Future robotic and manned missions to Mars with a significant increase in landed mass and a corresponding increase in entry heating will require the certification of heritage or new materials in relevant environments. Currently European<sup>8-11</sup>, Russian<sup>12</sup>, and Japanese<sup>13</sup> facilities are performing significant research using CO<sub>2</sub> as a test gas. NASA Langley Research Center's Hypersonic Materials Environmental Test System (HYMETS) facility was recently modified to provide a CO<sub>2</sub> test environment, and to serve as a pilot facility for CO<sub>2</sub> enhancements to the larger arc-jet facility at NASA Johnson Space Center. The purpose of this paper is to describe the modification, operation, and performance of the HYMETS arc-jet facility using CO<sub>2</sub> as a test gas. A basic comparison of heat fluxes, bulk properties, and performance characteristics for various Earth and Martian entry environments in HYMETS is provided. The Earth and Martian entry environments consist of a standard Earth atmosphere, an oxygen-rich Earth atmosphere, and a simulated Martian atmosphere. Finally, a preliminary comparison of the HYMETS arc-jet facility to several European plasma facilities is made.

## II. HYMETS Facility

### A. Facility History

The HYMETS facility was installed at NASA Langley Research Center in 1968 as a 100 kW segmented-constrictor-direct-current-electric-arc-heated plasma wind tunnel. Throughout the 1970's, 80's, 90's, and early 2000's it was used primarily for emissivity, catalysis, and dynamic oxidation testing of metals and coatings for hypersonic vehicles.<sup>14-16</sup> The range of test conditions for the HYMETS facility during that time is presented in Table 1. The facility was upgraded to 400 kW in 2005 expanding its range of test conditions to those presented in Table 2.<sup>17</sup> Since then HYMETS has been used primarily for characterization of ceramic matrix composite (CMC) materials, rigid and flexible ablators, high-temperature coatings, and for performing research and development on plasma flow diagnostics.

**Table 1. HYMETS Historic Test Conditions**

Specimen Surface Temperature (°F)	1472 – 2732
Specimen Stagnation Pressure (atm)	0.004 – 0.008
Free Stream Mach Number	3.5
Free Stream Enthalpy (Btu/lb <sub>m</sub> )	1719 – 4730
Cold Wall Heating Rate (Btu/ft <sup>2</sup> .s)	70 – 400

**Table 2. HYMETS Current Test Conditions**

Specimen Surface Temperature (°F)	2300 – 4500
Specimen Stagnation Pressure (atm)	0.013 – 0.079
Free Stream Mach Number	5.0
Free Stream Enthalpy (Btu/lb <sub>m</sub> )	2300 – 11500
Cold Wall Heating Rate (Btu/ft <sup>2</sup> .s)	100 – 600

### B. Facility Configuration

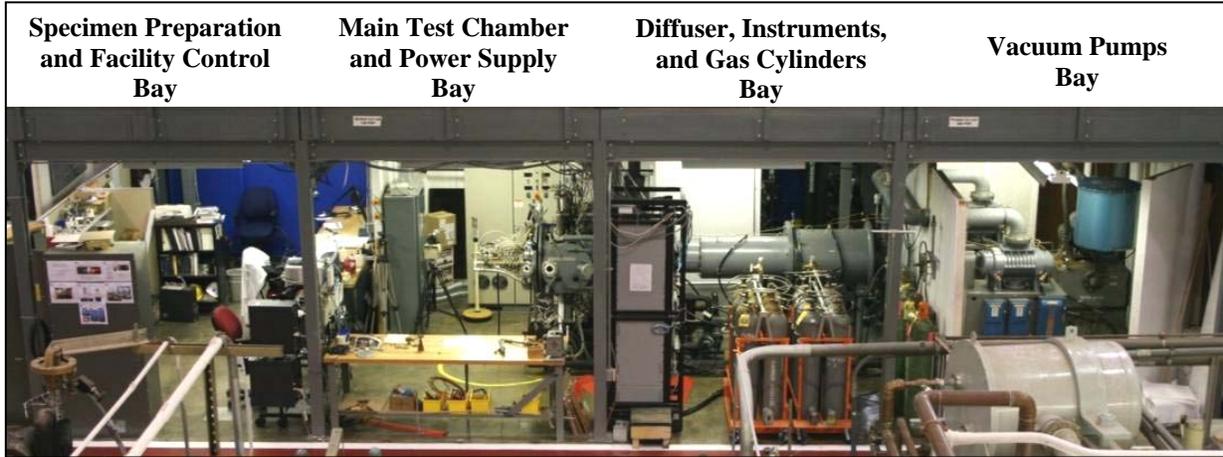
The HYMETS facility is shown in its entirety in Fig. 1. HYMETS is configured with a segmented-constrictor-dc-electric-arc-heater as an arc plasma generator. The arc plasma generator, schematic and photograph shown in Figs. 2 and 3 respectively, is mounted on the outside of the test chamber door. It consists of a copper cathode with tungsten button emitter, 32 electrically-isolated copper segment constrictors with a 0.5-inch diameter bore, a copper divergent-ring anode, and a copper convergent-divergent Mach 5 nozzle. The nozzle has a 0.5-inch diameter throat, 2.5-inch diameter exit plane, and a half-angle of 8 degrees. All arc plasma generator components are water cooled with jackets or by internal passageways.

Test gasses are injected tangentially into the bore of the arc plasma generator at six discrete locations. The gasses are heated by a high-voltage electric arc that is maintained between the cathode and anode to create a high temperature ionized plasma flow. The electric arc is spin-stabilized in the arc plasma generator by the vortex motion of the injected test gasses. The test gasses used in the arc plasma generator are supplied by several compressed gas cylinders, shown in Fig. 4, and can be custom mixed to the desired atmosphere. Adjustable volume percentages of N<sub>2</sub> and Argon (Ar) are used as shield gasses near the cathode and anode, respectively, to protect the electrodes from rapid oxidation. Power to the electrodes is provided by a 400 kW power supply with an induction filter shown in Fig. 5.

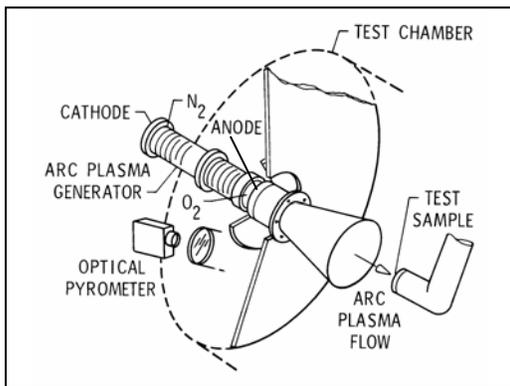
The plasma flow from the arc plasma generator is accelerated through the nozzle and exhausted into a 2-foot diameter by 3-foot long vacuum test chamber where it stagnates on one of four water-cooled injection stings arranged symmetrically around the inside circumference of the test chamber. The flow is then captured by a collector cone with an 8-inch diameter inlet plane, a 6-inch diameter constant cross-section diffuser, and a coiled-copper tubing heat exchanger to decelerate and cool the flow. A two-stage, continuous-flow, high-mass-capacity, mechanical pumping system, shown in Fig. 6, is used to evacuate the plasma flow from the facility. The collector, diffuser, heat exchanger, and mechanical pumping system provide for expansion of the plasma flow from the nozzle. The facility is cooled by a 150 ton re-circulating chiller with associated booster pumps and heat exchangers.

The HYMETS facility is computer-controlled by commercially available software<sup>18</sup> from a PC workstation and a Programmable Logic Control (PLC) touch-screen module shown in Fig. 7. Test conditions are controlled by adjusting the current and test gas mass flow rate setpoints for the arc plasma generator, which can be operated either

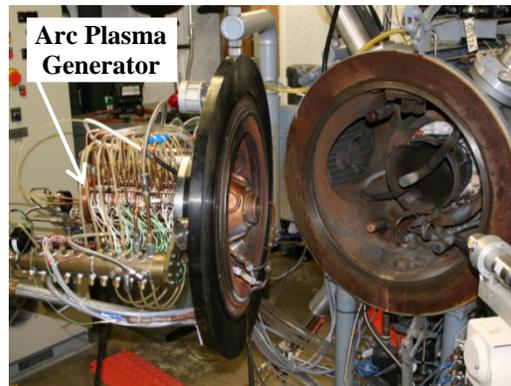
manually or automatically allowing for the generation of test profiles. Test specimens and instrumentation probes are mounted on the four injection stings which are inserted into the plasma flow by the PLC touch-screen module.



**Figure 1. HYMETS Facility**



**Figure 2. HYMETS Facility Test Setup Schematic**



**Figure 3. HYMETS Facility Test Setup**



**Figure 4. Test Gas Cylinders**



**Figure 5. Power Supply and Induction Filter**



**Figure 6. Mechanical Pumping System**



**Figure 7. HYMETS Facility Control Software and Touch-Screen Module**

### C. Facility Instrumentation

A wide variety of instrumentation is used to determine the flow characteristics of the plasma in an arc-jet facility.<sup>19</sup> Instrumentation used in the HYMETS facility consists of a Pitot tube which measures stagnation pressure, a SiC probe<sup>20</sup> which measures semi-catalytic hot-wall heat flux, a Gardon Gauge<sup>21</sup> and a Copper Slug Calorimeter<sup>22,23</sup> which measure fully-catalytic cold-wall heat flux, and a Teflon<sup>®</sup> Slug Calorimeter<sup>24,25</sup> which measures non-catalytic cold-wall heat flux. The Gardon Gauge and the Pitot tube, shown in Figs. 8 and 9 respectively, are located on the two lower injection stings inside the test chamber. The Copper Slug Calorimeter, shown in Fig. 10, is located on one of two upper injection stings inside the test chamber. The Teflon<sup>®</sup> Slug Calorimeter, SiC probe, and a test specimen configuration are shown in Figs. 11, 12, and 13, respectively. Any one of these items can be located on the remaining upper injection sting. The standard geometry for HYMETS instrumentation consists of a 1.3-inch diameter flat-faced cylinder with a 0.125-inch edge radius. The test specimen configuration shown in Fig. 13 is of the same basic geometry, but includes a 0.9-inch-diameter 0.03-inch-deep recess on the flat-face of the cylinder due to the lip on the specimen holder. The bulk enthalpy of the plasma flow is determined by performing an energy balance on the arc plasma generator.<sup>26</sup> A two-color (ratio) pyrometer and full-color digital video camera with variable exposure settings positioned outside the test chamber are remotely aimed at the specimen through a viewport on the test chamber door. The stagnation pressure, various heat fluxes, temperature, bulk enthalpy, and other facility parameters are recorded at 1000 Hz. The test results in this paper focus on the bulk enthalpy calculation and the measurements made with the Pitot tube and the Copper and Teflon<sup>®</sup> Slug Calorimeters.

An energy balance is performed on the arc plasma generator in HYMETS to calculate the bulk enthalpy. The energy input to the arc plasma generator is determined by the product of voltage and current measurements. The energy removal from the arc plasma generator is determined by the product of the cooling water flow rate and the differential temperature measured across the inlet and outlet of the cooling water manifolds. Finally, the energy that remains in the arc plasma generator is divided by the total measured mass flow rate of the test gases. The bulk enthalpy is calculated using<sup>26</sup>

$$H_o = \left( \frac{E I C_1 - C_2 CW C_p \rho \Delta T_{CW}}{M_{Total}} \right) \quad (1)$$

where the constants  $C_1$  and  $C_2$  convert the energy input and output to the proper units respectively.

The Copper Slug Calorimeter used in HYMETS to determine fully-catalytic cold-wall heat flux consists of an un-cooled slug sensor element that is 0.5-inches in diameter by 0.5-inches-long with an un-cooled shroud that is 1.3-inches in diameter by 0.850-inches-long and a flow-face edge radius of 0.125-inches. The slug sensor element and shroud are fabricated out of oxygen-free high-conductivity (OFHC) copper. The slug sensor element has a 0.002-inch-wide “insulating” air gap between it and the shroud, and is held in place using six cone-tipped set-screws. The slug sensor element also has a Type-K thermocouple mounted on its back surface to measure temperature rise. The length, diameter, and mass of the slug sensor element are measured prior to testing. The Copper Slug Calorimeter is inserted into a steady-state flow for 2-3 seconds so that it achieves a back-face temperature rise of several hundred

degrees Fahrenheit, not to exceed a final temperature of 600°F. The fully-catalytic cold-wall heat flux is calculated from the density of the OFHC copper slug sensor element, the temperature dependent specific heat capacity of the element, the length of the element, and the slope of the temperature rise from the linear portion of the temperature response curve for the Type-K thermocouple using<sup>22</sup>

$$\dot{q}_{FCCW} = \rho C_p(T) l \left( \frac{\Delta T_{FCCW}}{\Delta t} \right) \quad (2)$$

The uncertainty of the resulting method given by Eq. (2) is assumed to be +/-10%.<sup>23</sup>

The Teflon<sup>®</sup> Slug Calorimeter used in HYMETS to determine non-catalytic cold-wall heat flux consists of a slug sensor element with a shroud made from virgin Teflon<sup>®</sup> PTFE (poly-tetra-fluoro-ethylene). Teflon<sup>®</sup> PTFE is a non-catalytic, low-temperature, subliming ablator. The design of the slug sensor element and shroud are similar to the design for the Copper Slug Calorimeter without positioning set-screws and the Type-K thermocouple. The slug sensor element has an interference fit with the shroud so that it is easily assembled and disassembled. The length, diameter, and mass of the slug sensor element are measured before and after testing. The Teflon<sup>®</sup> Slug Calorimeter is inserted into a steady-state flow for 30 seconds according to Refs. 24 and 25. The non-catalytic cold-wall heat flux is calculated from the change in mass of the slug sensor element, the amount of time it was exposed to the flow, the effective hemispherical radius of the slug sensor element, the flat-face radius of the element, and the stagnation pressure at the steady-state flow condition using<sup>25</sup>

$$\dot{q}_{NCCW} = \left[ \frac{\dot{m}_T R_{eff}^{0.21}}{0.0046 P_{stag}^{0.24}} \right]^{(1/0.55)} \quad (3)$$

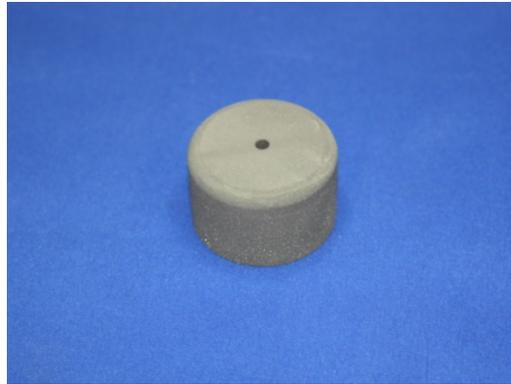
where

$$R_{eff} = 3.3R_{FlatFace} \quad (4)$$

Equation (3) has an assumed uncertainty of +/-10%, and is based on the correlation of an extensive collection of arc-jet data on Teflon<sup>®</sup> PTFE for a variety of geometries and flow conditions.<sup>24,25</sup>



**Figure 8. Gardon Gauge**



**Figure 9. Pitot Tube**



Figure 10. Copper Slug Calorimeter

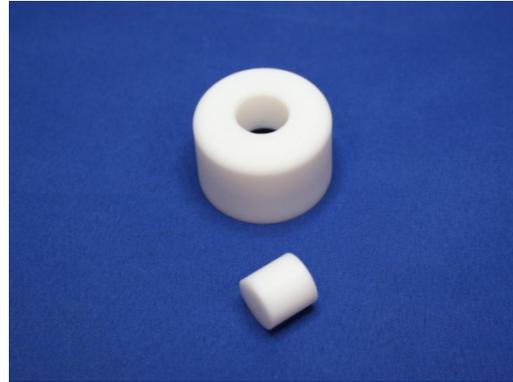


Figure 11. Teflon® Slug Calorimeter



Figure 12. SiC Probe

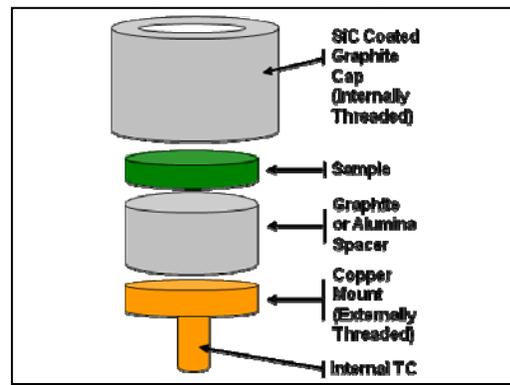


Figure 13. Specimen Holder Configuration

#### D. Facility CO<sub>2</sub> Modification and Operation

The implementation of CO<sub>2</sub> as a test gas in the HYMETs facility was performed incrementally. The first step was to add a small percentage of CO<sub>2</sub> gas into the arc plasma generator to the N<sub>2</sub>, O<sub>2</sub>, and Ar gasses already present. The percentage of CO<sub>2</sub> gas was then increased, simultaneously decreasing the percentages of N<sub>2</sub> and O<sub>2</sub>, while maintaining electric arc stability and arc plasma generator integrity. The current operational CO<sub>2</sub> gas ratio for HYMETs is 24% N<sub>2</sub>, 71% CO<sub>2</sub>, and 5% Ar by volume. This provides a 25% to 75% ratio by volume of N<sub>2</sub> to CO<sub>2</sub> when Ar is excluded from the mix. This current ratio is closer to achieving the actual gas ratio for the Martian atmosphere of 3% N<sub>2</sub> to 97% CO<sub>2</sub> than historical attempts. For reference, the operational air ratio for HYMETs is 75% N<sub>2</sub>, 20% O<sub>2</sub>, and 5% Ar by volume. This provides a 79% to 21% ratio by volume of N<sub>2</sub> to O<sub>2</sub> when Ar is excluded from the mix, which is the actual gas ratio for the Earth atmosphere. The arc plasma generator injection locations for both air and CO<sub>2</sub> operation are shown in Fig. 14. A fixed amount of N<sub>2</sub> was retained near the cathode in the arc plasma generator to prevent oxidation. CO<sub>2</sub> was supplied from a compressed gas cylinder, which contained both gaseous and liquid CO<sub>2</sub> at room temperature. It was noted that during sustained operation of the facility with CO<sub>2</sub>, liquid CO<sub>2</sub> boiled off in the cylinder to satisfy the gaseous CO<sub>2</sub> demand. The heat removed from the cylinder to boil off the liquid created a temperature drop in the gas. When the low temperature CO<sub>2</sub> gas expanded through the pressure regulator for the CO<sub>2</sub> cylinder, dry-ice formed in the regulator blocking the flow of gas through the distribution lines. A tube and shell heat exchanger for the distribution lines with three pressure-regulated stages was created to prevent this formation of dry-ice. Even with the heated distribution lines and regulators, it was still possible for dry-ice to form inside the cylinder if a sufficient volume of gas was removed over a short time. To avoid removing an excessive volume of gas, CO<sub>2</sub> was supplied from eight cylinders simultaneously. It was also noted that sustained operation of the facility with CO<sub>2</sub> introduced a “soot-like” deposit onto the components of the arc plasma generator. The build-up of the “soot-like” deposits decreased the stability of the electric arc and increased the risk of its attachment to a segment resulting in damage to the arc plasma generator. The “soot-like” deposits require regular cleaning of the arc segments with a wire brush.

The second step in implementing CO<sub>2</sub> was to assess the industrial and environmental safety implications associated with mixing CO<sub>2</sub> and N<sub>2</sub> because of the potential to produce harmful molecules like Carbon Monoxide (CO), Cyanide (CN), and Nitric Oxide (NO). The production of CO and O<sub>2</sub> creates a possible explosion hazard within the heat exchanger and mechanical pumping system, as CO is a fuel. The upper explosive limit and lower explosive limit as provided by the Material Safety Data Sheet (MSDS) for CO in air is 74% and 12.5% respectively. A residual gas analysis (RGA) of exhaust gasses in the section between the heat exchanger and mechanical pumping system performed by Sandia National Laboratories revealed that half of the CO<sub>2</sub> injected into the arc plasma generator was disassociated into its constituents.<sup>27</sup> If these constituents were assumed to be entirely composed of CO and O<sub>2</sub>, then CO would make up 35.6% of the heat exchanger and mechanical pumping system gasses, which is within the explosive limit. However, CO and O<sub>2</sub> mixtures are not combustible below about 25 torr<sup>28</sup>, and the maximum pressure in the heat exchanger and mechanical pumping system is 25.4 torr. To further mitigate the explosive mix of CO and O<sub>2</sub> in the heat exchanger and mechanical pumping system, the facility is purged during start-up and shutdown using only the inert gasses N<sub>2</sub> and Ar when operating with CO<sub>2</sub>.<sup>29</sup>

The production of CN during facility operation with CO<sub>2</sub> and N<sub>2</sub> would create a hazardous and undesirable work environment as CN is a poison. Fortunately RGA measurements indicated no detectable levels of particulate CN or Hydrogen Cyanide (HCN) in the facility's exhaust.<sup>27</sup> Additional surface swabs and air samples taken inside the test chamber before and after facility operation also indicated no detectable levels of particulate CN or HCN.<sup>30</sup> Particulate CN was found in the pump oil for the mechanical pumping system.<sup>31</sup> The lack of particulate CN or HCN in the test chamber samples and RGA measurements upstream from the mechanical pumps indicates a different method of particulate CN creation than from the arc plasma generator. The particulate CN in the mechanical pumps could have formed through a reaction between the hydrocarbon pump oil and the hot (200°F), partially disassociated nitrogen in the exhaust gasses moving through it. Industrial safety personnel noted that because the particulate CN is entrapped in pump oil, which itself is considered a hazardous material, the standard method for handling and disposal was sufficient.

Finally, the production of NO during facility operation with CO<sub>2</sub> and N<sub>2</sub> means that HYMETs would have to be considered a pollution source. Nitric Oxide production was confirmed by RGA measurements<sup>27</sup> and NO Planar Laser Induced Fluorescence (PLIF) measurements.<sup>32</sup> Emission of NO from the HYMETs facility is currently regulated as part of the NASA Langley air permit for operation with N<sub>2</sub> and O<sub>2</sub>.<sup>33</sup> There is sufficient margin in the HYMETs portion of the Langley air permit to absorb the additional NO that is produced by operation with CO<sub>2</sub> and N<sub>2</sub>. No further restrictions were required for operation with CO<sub>2</sub> and N<sub>2</sub> since the emission of CO<sub>2</sub>, N<sub>2</sub>, and their constituents (i.e. CO, CN, C, N, O<sub>2</sub>, and O) are not regulated in the Commonwealth of Virginia.

The remaining step is to better simulate the Martian atmosphere by increasing the operational gas ratio of CO<sub>2</sub> to 97% CO<sub>2</sub> to 3% N<sub>2</sub> by volume. Higher ratios of CO<sub>2</sub> to N<sub>2</sub> than the current ratio, 75% CO<sub>2</sub> to 25% N<sub>2</sub> by volume, result in unacceptable oxidation of the cathode. A possible method of achieving higher ratios is to introduce CO near the cathode to reduce the amount of N<sub>2</sub> needed to shield the cathode. The tungsten button in the cathode is not easily oxidized by CO.<sup>34</sup> The CO would then be balanced by introducing a chemically equivalent amount of O<sub>2</sub> closer to the anode. In addition to bringing the test gas ratio closer to that of the actual Martian atmosphere, the facility would be able to take advantage of the heat of formation of CO and O<sub>2</sub> in the production of CO<sub>2</sub> to increase the bulk enthalpy and heat flux of the test conditions. An analysis determined this would be the best way to achieve MSL-like entry conditions without significant modifications to the facility.<sup>35</sup> Since CO is both a fuel and a poison, an industrial safety study determined that the CO would have to be located in an open atmosphere outside of the facility's building and piped in for testing. The time and cost associated with building a cylinder storage area and gas distribution system for CO prevented its use in this study.

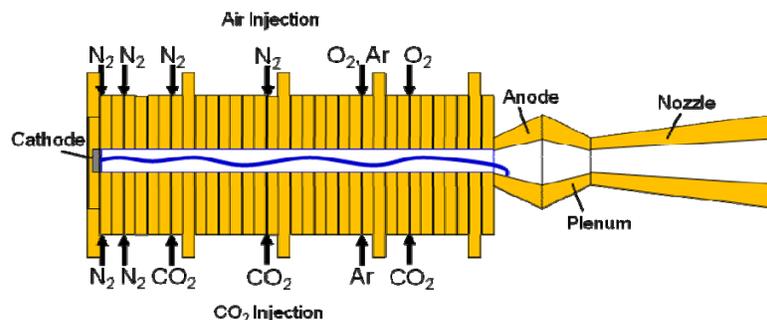


Figure 14. HYMETs Arc Plasma Generator Test Gas Injection Schematic

### III. Test Approach

#### A. Test Conditions

A range of test conditions, or envelope map, for the HYMETs facility was performed with a standard and oxygen-rich Earth atmosphere and a simulated Martian atmosphere as defined by the test gas constituents listed in Table 3. The parameters that can be controlled when operating the facility are the arc current and the test gas mass flow rate. A particular combination of these parameters, a so-called facility setpoint, yields a test environment with a certain bulk enthalpy, heat flux, and stagnation pressure, which are the desired parameters for materials testing. An envelope map consists of running the facility at a number of setpoints to determine the range of achievable parameters. A complete matrix of the HYMETs facility setpoints is listed in Table 4. The current and mass flow rate setpoints higher than 300 Amps in Table 4 were unavailable at the time of this testing because of an instability within the power supply which limited its maximum current output. The power supply instability has since been fixed, and these setpoints are now available for future testing. The number of setpoints at which tests could be conducted was further reduced by limited availability of Teflon<sup>®</sup> Slug Calorimeters. The tests in this paper were conducted at the gray highlighted setpoints listed in Table 4. Facility operating conditions including bulk enthalpy, heat flux, stagnation pressure, mass flow rate, arc current, and arc voltage were recorded at all times during facility operation. Results from these tests were used to develop comparisons between enthalpies, heating rates, and stagnation pressures for the three different test atmospheres. Of particular interest is a comparison of results between the oxygen-rich Earth atmosphere and the simulated Martian atmosphere to assess the adequacy of testing in an oxygen-rich Earth atmosphere as a replacement for testing with CO<sub>2</sub>. The oxygen-rich Earth atmosphere in Table 3 contains 27.6% O<sub>2</sub> by weight, which is equivalent to the simulated Martian atmosphere which can generate 27.3% O<sub>2</sub> by weight from the disassociation of CO<sub>2</sub>.<sup>4</sup>

**Table 3. HYMETs Test Atmospheres**

Test Atmosphere	Test Gases				Measurement Basis
	N <sub>2</sub>	O <sub>2</sub>	CO <sub>2</sub>	Ar	
Standard Earth Atmosphere	75%	20%	---	5%	By Volume (or Mole)
Oxygen-Rich Earth Atmosphere	71%	24%	---	5%	By Volume (or Mole)
Simulated Martian Atmosphere	24%	---	71%	5%	By Volume (or Mole)

**Table 4. HYMETs Facility Setpoint Matrix for Test Atmospheres**

Current (Amps)	Mass Flow Rate (lb <sub>m</sub> /s)										
	0.0048	0.0072	0.0096	0.0120	0.0145	0.0169	0.0193	0.0217	0.0241	0.0265	0.0289
100	0.0048	0.0072	0.0096	0.0120	0.0145	0.0169	0.0193	0.0217	0.0241		
150	0.0048	0.0072	0.0096	0.0120	0.0145	0.0169	0.0193	0.0217	0.0241		
200	0.0048	0.0072	0.0096	0.0120	0.0145	0.0169	0.0193	0.0217	0.0241		
250		0.0072	0.0096	0.0120	0.0145	0.0169	0.0193	0.0217	0.0241	0.0265	
300		0.0072	0.0096	0.0120	0.0145	0.0169	0.0193	0.0217	0.0241	0.0265	
350			0.0096	0.0120	0.0145	0.0169	0.0193	0.0217	0.0241	0.0265	0.0289
400			0.0096	0.0120	0.0145	0.0169	0.0193	0.0217	0.0241	0.0265	0.0289

#### B. Test Data

An effort was made to reduce the precision and bias uncertainties in the test results to a 95% confidence interval. The precision uncertainty was reduced by using a median filter processing technique to remove spurious noise from the recorded parameters. The median filter<sup>18</sup> consists of a data point bounding interval which is set by the left and right rank, or measurement points. The values within the interval bounded by the rank are arranged in ascending numerical order and the median value is found. The filtered curve is the median value (middle-most point) for the corresponding interval. Given the noise characteristics of the HYMETs parameters, a left and right rank of 2 measurement points was selected. The bias uncertainty for the filtered parameters was determined by the accuracy of the transducer. These values are typically reported as a percentage of the instrument's full scale range or as a percentage of the reading with a 95% confidence level. The precision and bias uncertainties were then combined to determine the total uncertainty for each parameter.

### C. European Test Conditions

Additional tests were performed in HYMETS with test conditions replicating those from an experimental study of several European plasma facilities presented in Refs. 36 and 37. The sensors used in the European study consisted of SiC probes, Gardon Gauges, Copper Calorimeter Probes, and Heat Flux Micro-sensors. The study utilized three standard model geometries. The third geometry in the study, a 1.97-inch flat-faced cylinder with 0.45-inch-edge-radius, was the most similar to HYMETS. The European plasma facilities that tested with this geometry included the segmented-arc-heated L3K facility at the German Aerospace Center (DLR)<sup>8</sup>, the thermal or magneto-plasma-dynamic Plasma Wind Tunnels (PWK) 1 and 2 at the Institute of Space Systems at the University of Stuttgart (IRS)<sup>9</sup>, the Huels arc-heated SIMOUN and inductively-heated COMETE facilities at the European Aeronautic Defense and Space Company (EADS)<sup>10</sup>, and the inductively-heated PLASMATRON facility at the von Karman Institute (VKI)<sup>11</sup>. The representative and comparable standard Earth atmosphere flow test conditions are listed in Table 5. The stagnation pressure, bulk enthalpy, and heat flux measurements made in the HYMETS facility at these flow test conditions were compared to those made in the European facilities.

**Table 5. HYMETS Flow Test Conditions for Comparison with European**

Test Conditions	Stagnation Pressure	Bulk Enthalpy
Flow Test Condition I (FTC-I)	0.0345 atm (35 hPa)	3870 Btu/lb <sub>m</sub> (9 MJ/kg)
Flow Test Condition II (FTC-II)	0.0345 atm (35 hPa)	5589 Btu/lb <sub>m</sub> (13 MJ/kg)

## IV. Test Results

The effect of the test atmosphere on facility performance is assessed by comparing the bulk enthalpy, various measured heat flux, and stagnation pressure at the same facility setpoint for the three simulated entry environments. All parameters are displayed with respect to the mass flow rate to provide a clear relationship between the facility setpoints and facility performance. A comparison of test results for a standard Earth atmosphere and simulated Martian atmosphere is presented. A comparison of test results for an oxygen-rich Earth atmosphere and the simulated Martian atmosphere is also presented. These comparisons are made to assess the adequacy of testing in a standard or an oxygen-rich Earth atmosphere as a replacement for testing with CO<sub>2</sub>. In addition to the comparisons, the results presented in the following subsections establish the current range of test conditions for the HYMETS facility.

### A. Comparison of Simulated Martian and Standard Earth Atmosphere

A representative image of the plasma flow for the standard Earth atmosphere is shown in Fig. 15 where the plasma exhibits a pinkish-orange glow. A representative image of the plasma flow for the simulated Martian atmosphere is shown in Fig. 16 where the plasma exhibits a whitish-blue glow. The constituents in the plasma determine its color. Research to determine the constituents of the plasma in HYMETS will be described in a later paper.

Bulk enthalpy is a parameter used to characterize the flow environment when comparing performance of different materials and for extrapolating test results to flight. The comparison of the bulk enthalpy for the standard Earth and simulated Martian atmospheres in Fig. 17 shows that the simulated Martian bulk enthalpy values are higher than the standard Earth values for each of the facility setpoints. Examining the variables that go into Eq. (1) for the calculation of the bulk enthalpy, the current and mass flow rates for each atmosphere are identical per the test conditions. The arc plasma generator voltage is 4.4% more for the simulated Martian atmosphere than it is for the standard Earth atmosphere based on the root-mean-square (RMS) of the percent differences between the two. The RMS value was determined by first taking the percent difference of a Martian value from an Earth value at each of the facility setpoints for a given parameter. These individual setpoint percent differences were then combined into one value for that parameter using the RMS method, in an effort to provide a single measure of the overall difference between the two atmospheres. The energy removed by the cooling water from the arc plasma generator is 21.7% less for the simulated Martian atmosphere than it is for the standard Earth atmosphere based on the RMS of the percent differences between the two. The contributions from these two variables create the offset in the bulk enthalpies between the simulated Martian and standard Earth atmospheres. The difference in voltage can be attributed to the differences in the ionization and disassociation energies for each of the test gasses that make up the

two atmospheres. The reason for the greater energy loss to the cooling water for the standard Earth atmosphere over the simulated Martian atmosphere is unknown at this time. The RMS percent difference with the maximum and minimum of the individual setpoint percent differences between the two atmospheres for the bulk enthalpy are presented in Table 6. The RMS percent total uncertainty with the maximum and minimum of the individual setpoint percent total uncertainties in the bulk enthalpy measurement are also presented in Table 6 for each atmosphere. The total uncertainties for the standard Earth and simulated Martian atmospheres are similar because the measurements for the two atmospheres were made using the same instruments and data acquisition system.

Heat flux is a parameter chosen to establish the environment in which a material must perform. Two types of cold-wall heat flux measurements were made. The fully-catalytic cold-wall heat flux is measured by the Copper Slug Calorimeter. The non-catalytic cold-wall heat flux is measured by the Teflon<sup>®</sup> Slug Calorimeter. The catalytic energy, or fully-catalytic heat flux minus the non-catalytic heat flux, is the energy contributed by the recombination of the disassociated gas constituents when they come in contact with a catalytic surface. The non-catalytic heat flux provides a measure of the heat energy in the plasma. The fully-catalytic heat flux provides a measure of the total energy in the plasma, that is the heat energy plus the catalytic energy. Comparisons of the fully-catalytic and non-catalytic heat flux measurements for the two atmospheres are provided in Figs. 18 and 19, respectively. The heat flux measurements for the simulated Martian atmosphere at the 100 Amp current setpoint are lower than the standard Earth atmosphere for each of the mass flow rates. At the 200 Amp current setpoint, the heat flux measurements for the simulated Martian atmosphere are higher than the standard Earth atmosphere for the lower mass flow rates, but then become lower than the standard Earth atmosphere as the mass flow rates increase. Finally, the heat flux measurements for the simulated Martian atmosphere at the 300 Amp current setpoint are higher than the standard Earth atmosphere for each of the mass flow rates. The RMS percent difference with the maximum and minimum of the individual setpoint percent differences between the two atmospheres for the fully-catalytic and non-catalytic heat flux measurements are presented in Table 6. The RMS percent total uncertainty with the maximum and minimum of the individual setpoint percent total uncertainties in these measurements for each atmosphere are also presented in Table 6. A comparison of the percentages of the catalytic energy in the plasma for the two atmospheres is provided in Fig. 20. The standard Earth atmosphere appears to have more catalytic energy for each of the facility setpoints than the simulated Martian atmosphere. The reason for this is unknown. However, the trends exhibited by the catalytic energy in Fig. 20 are expected. At low mass flow rates, and therefore low partial gas pressures, the plasma for the two atmospheres is highly disassociated contributing a higher amount of catalytic energy. As the mass flow rates increase, the partial pressure of each test gas increases reducing the amount of disassociation and decreasing the amount of catalytic energy.

Finally, the stagnation pressure is also a parameter chosen to establish the environment in which a material must perform. Achieving a stagnation pressure test condition is usually secondary to achieving a bulk enthalpy or heat flux test condition. The stagnation pressure is therefore optimized to be close to the desired test condition. A comparison of the stagnation pressures for the two atmospheres is provided in Fig. 21. The stagnation pressures vary almost linearly with mass flow rate for the two atmospheres. A similar trend to that found in the heat flux measurement comparisons is present in the stagnation pressures for the standard Earth and simulated Martian atmospheres at each of the current setpoints. The RMS percent difference with the maximum and minimum of the individual setpoint percent differences between the two atmospheres for the stagnation pressure are presented in Table 6. The RMS percent total uncertainty with the maximum and minimum of the individual setpoint percent total uncertainties in the stagnation pressure measurement are also presented in Table 6 for each atmosphere. The total uncertainties for the standard Earth and simulated Martian atmospheres are again similar because the measurements were made using the same instruments and data acquisition system. The low total uncertainties for the stagnation pressures and the scale of the y-axis make the error bars appear absent from Fig. 21.



**Figure 15. Standard Earth Atmosphere Plasma**



**Figure 16. Simulated Martian Atmosphere Plasma**

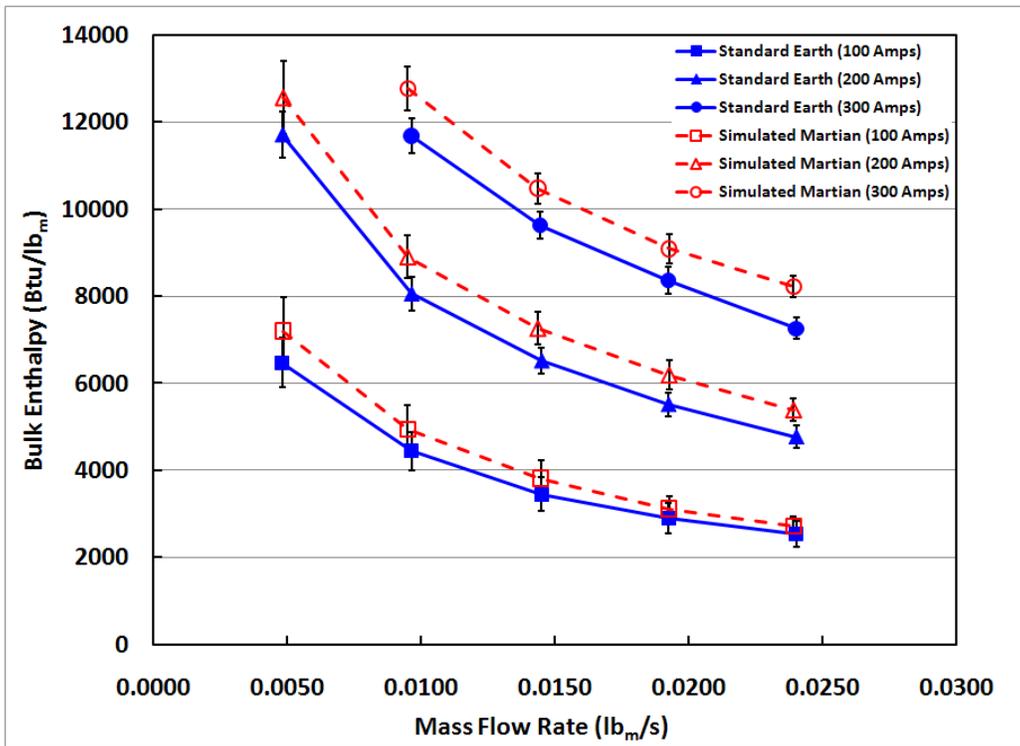


Figure 17. Bulk Enthalpy Comparison

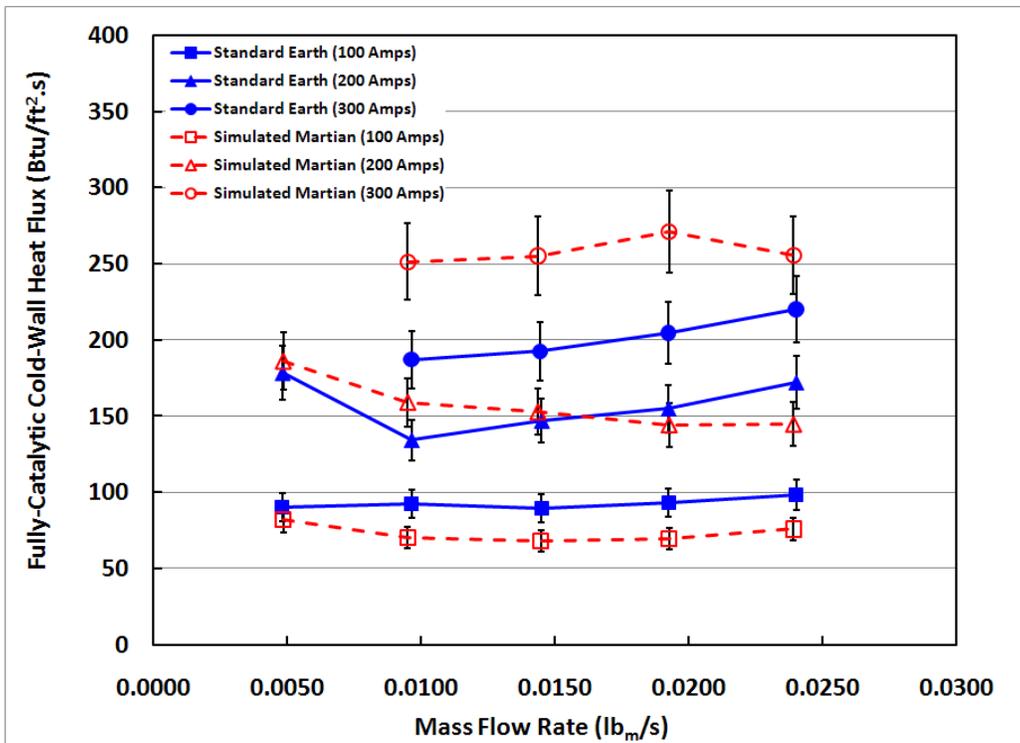


Figure 18. Copper Slug Calorimeter Cold-Wall Heat Flux Comparison

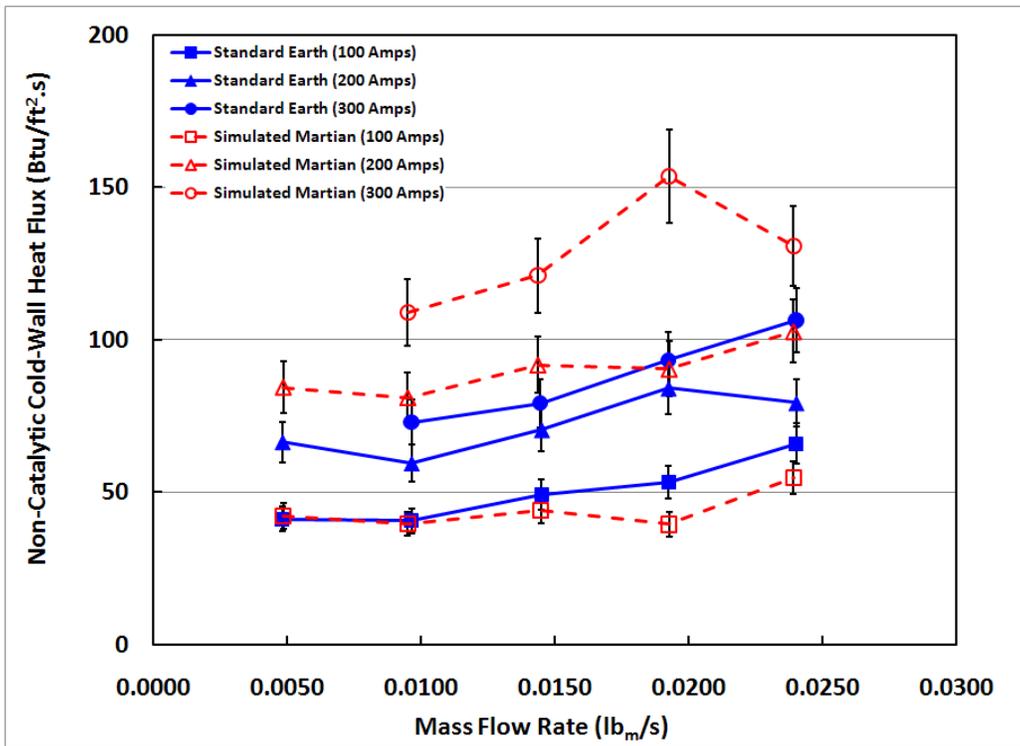


Figure 19. Teflon<sup>®</sup> Slug Calorimeter Cold Wall Heat Flux Comparison

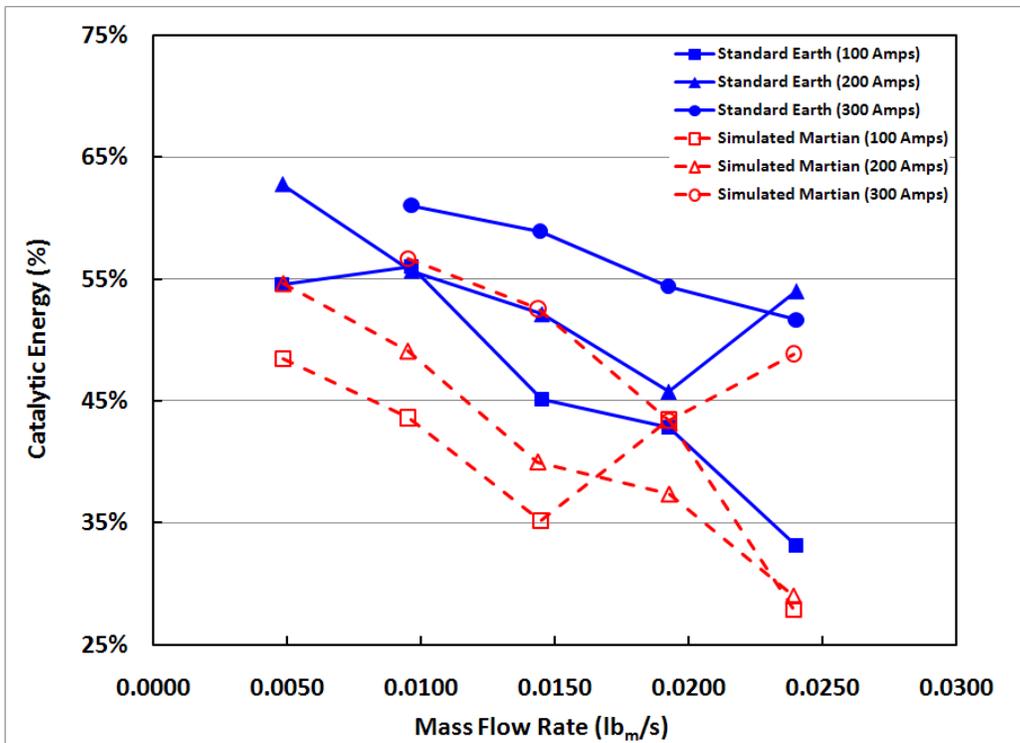


Figure 20. Catalytic Energy in the Plasma Comparison

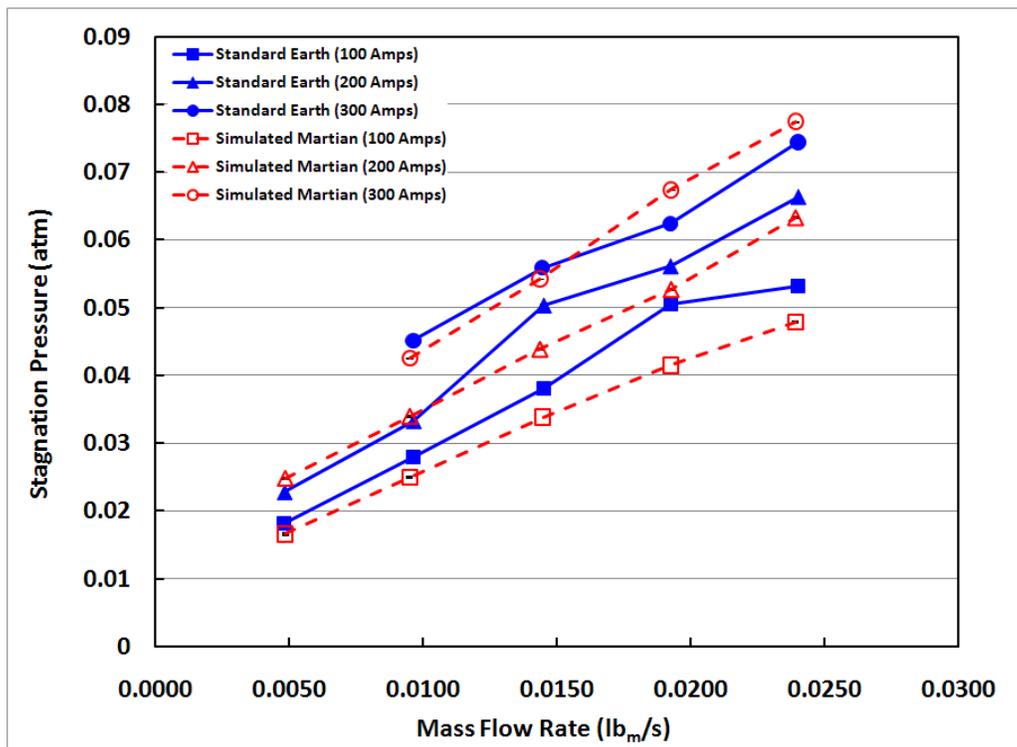


Figure 21. Stagnation Pressure Comparison

Table 6. Standard Earth and Simulated Martian Measurement Differences and Uncertainties

Bulk Enthalpy	RMS (%)	Maximum (%)	Minimum (%)
Stand. Earth to Sim. Martian Atm. Difference	10.7	13.2	7.0
Standard Earth Atmosphere Uncertainty	7.6	12.0	3.2
Simulated Martian Atmosphere Uncertainty	7.4	11.2	3.1

Fully-Catalytic Cold-Wall Heat Flux	RMS (%)	Maximum (%)	Minimum (%)
Stand. Earth to Sim. Martian Atm. Difference	21.6	34.2	-25.2
Standard Earth Atmosphere Uncertainty	10.4	10	-10
Simulated Martian Atmosphere Uncertainty	10.4	10	-10

Non-Catalytic Cold-Wall Heat Flux	RMS (%)	Maximum (%)	Minimum (%)
Stand. Earth to Sim. Martian Atm. Difference	33.8	64.5	-26.0
Standard Earth Atmosphere Uncertainty	10.4	10	-10
Simulated Martian Atmosphere Uncertainty	10.4	10	-10

Stagnation Pressure	RMS (%)	Maximum (%)	Minimum (%)
Stand. Earth to Sim. Martian Atm. Difference	12.7	32.4	-17.9
Standard Earth Atmosphere Uncertainty	0.21	0.63	0.08
Simulated Martian Atmosphere Uncertainty	0.26	0.81	0.09

## B. Comparison of Oxygen-Rich Earth Atmosphere and Simulated Martian Atmosphere

A representative image of the plasma flow for the oxygen-rich Earth atmosphere is shown in Fig. 22 where the plasma exhibits a similar color to the standard Earth atmosphere even with the extra oxygen. A representative image of the plasma flow for the simulated Martian atmosphere was shown in Fig. 16.

The comparison of the bulk enthalpy for the oxygen-rich Earth and simulated Martian atmospheres in Fig. 23 shows that the Martian bulk enthalpy values are again higher for each of the facility setpoints, with the values for the oxygen-rich Earth atmosphere being 1.8% higher than those of the standard Earth atmosphere based on the RMS of the percent differences between the two. The arc plasma generator voltage is 3.6% more for the simulated Martian atmosphere than it is for the oxygen-rich Earth atmosphere based on the RMS of the percent differences between the two. The energy removed by the cooling water from the arc plasma generator is 21.4% less for the simulated Martian atmosphere than it is for the oxygen-rich Earth atmosphere based on the RMS of the percent differences between the two. The RMS percent difference with the maximum and minimum of the individual setpoint percent differences between the two atmospheres for the bulk enthalpy are presented in Table 7. The RMS percent total uncertainty with the maximum and minimum of the individual setpoint percent total uncertainties in the bulk enthalpy measurement are also presented in Table 7 for each atmosphere. The total uncertainties for the oxygen-rich Earth and standard Earth atmospheres are identical.

Comparisons of the fully-catalytic and non-catalytic heat flux measurements for the oxygen-rich Earth and simulated Martian atmospheres are provided in Figs. 24 and 25, respectively. The heat flux measurement trends for the oxygen-rich Earth atmosphere when compared to the simulated Martian atmosphere are almost identical to those for the standard Earth atmosphere comparison. The oxygen-rich Earth atmosphere fully-catalytic cold-wall heat flux measurements are 6.0% higher than the standard Earth atmosphere measurements based on the RMS of the percent differences between the two. The oxygen-rich Earth atmosphere non-catalytic cold-wall heat flux measurements are 18.8% higher than the standard Earth atmosphere measurements based on the RMS of the percent differences between the two. The RMS percent difference with the maximum and minimum of the individual setpoint percent differences between the two atmospheres for the fully-catalytic and non-catalytic heat flux measurements are presented in Table 7. The RMS percent total uncertainty with the maximum and minimum of the individual setpoint percent total uncertainties in these measurements for each atmosphere are also presented in Table 7. A comparison of the percentages of the catalytic energy in the plasma for the oxygen-rich Earth and simulated Martian atmospheres is provided in Fig. 26. The catalytic energy for the oxygen-rich Earth atmosphere compares favorably with that of the simulated Martian atmosphere for each of the facility setpoints. This is in contrast to the same catalytic comparison made between the standard Earth and simulated Martian atmospheres. The reason for this is unknown. However, the result is encouraging when considering the substitution of an oxygen-rich Earth atmosphere for a simulated Martian atmosphere.

A comparison of the stagnation pressures for the oxygen-rich Earth and simulated Martian atmospheres is provided in Fig. 27. The stagnation pressures vary linearly with mass flow rate for the two atmospheres. The stagnation pressure trends for the oxygen-rich Earth atmosphere when compared to the simulated Martian atmosphere are almost identical to those for the standard Earth atmosphere comparison. The oxygen-rich Earth atmosphere stagnation pressures are 7.7% lower than the standard Earth atmosphere measurements based on the RMS of the percent differences between the two. The RMS percent difference with the maximum and minimum of the individual setpoint percent differences between the two atmospheres for the stagnation pressure are presented in Table 7. The RMS percent total uncertainty with the maximum and minimum of the individual setpoint percent total uncertainties in the stagnation pressure measurement are also presented in Table 7 for each atmosphere. The total uncertainties for the oxygen-rich Earth and standard Earth atmospheres are almost identical. The low total uncertainties for the stagnation pressures and the scale of the y-axis make the error bars appear absent from Fig. 27.



**Figure 22. Oxygen-Rich Earth Atmosphere Plasma**

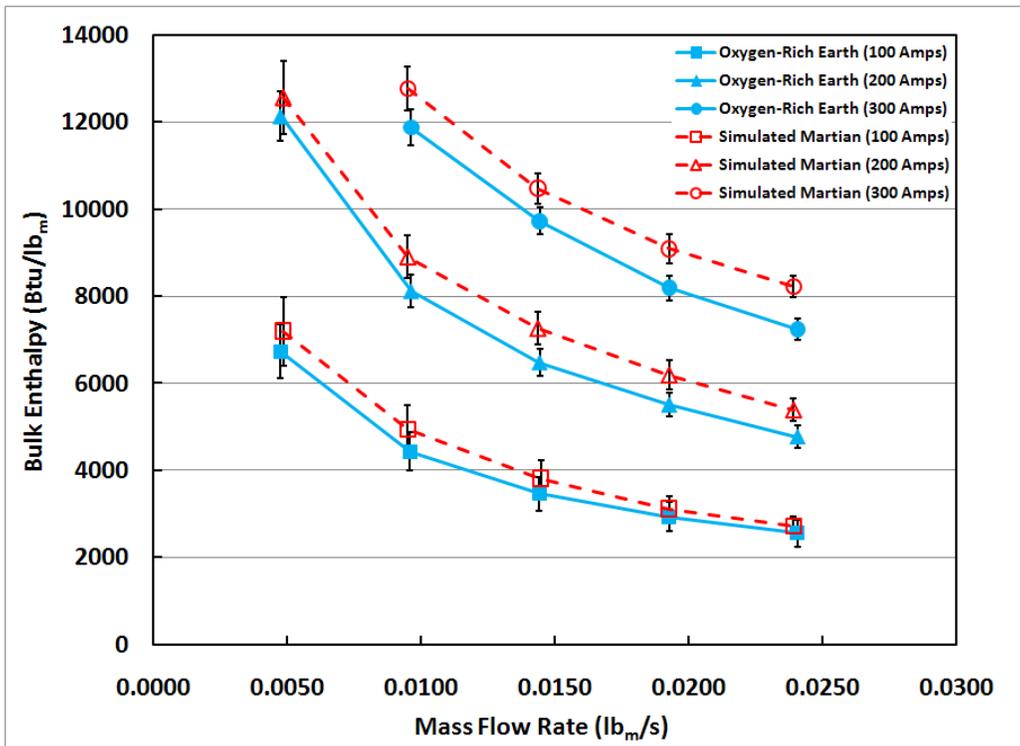


Figure 23. Bulk Enthalpy Comparison

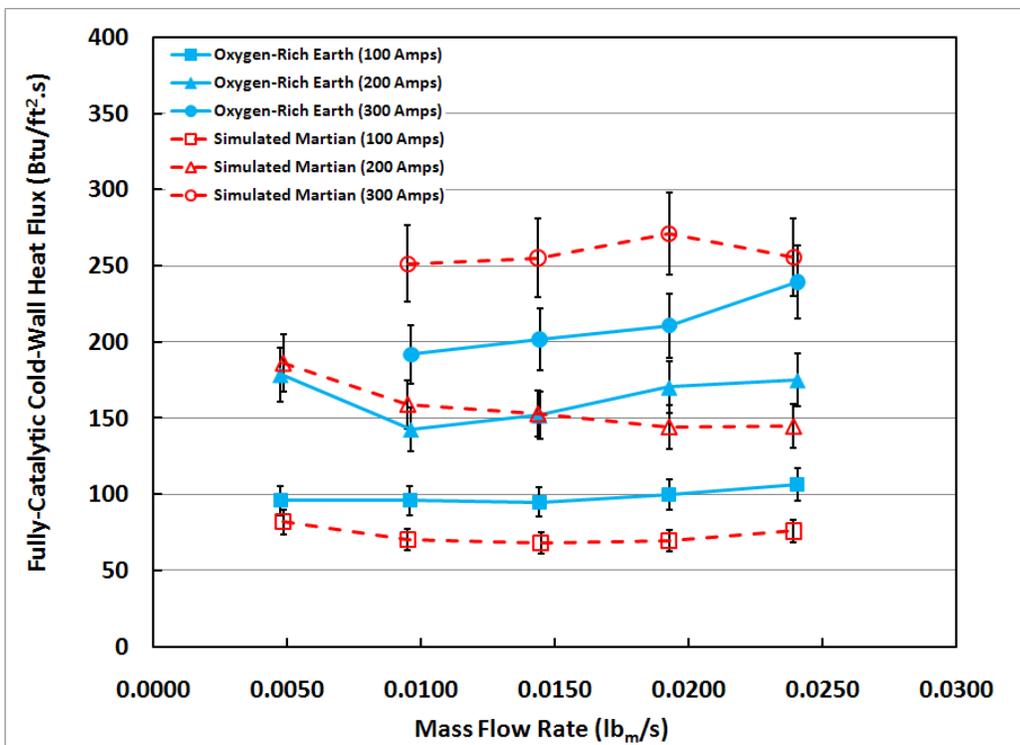


Figure 24. Copper Slug Calorimeter Cold Wall Heat Flux Comparison

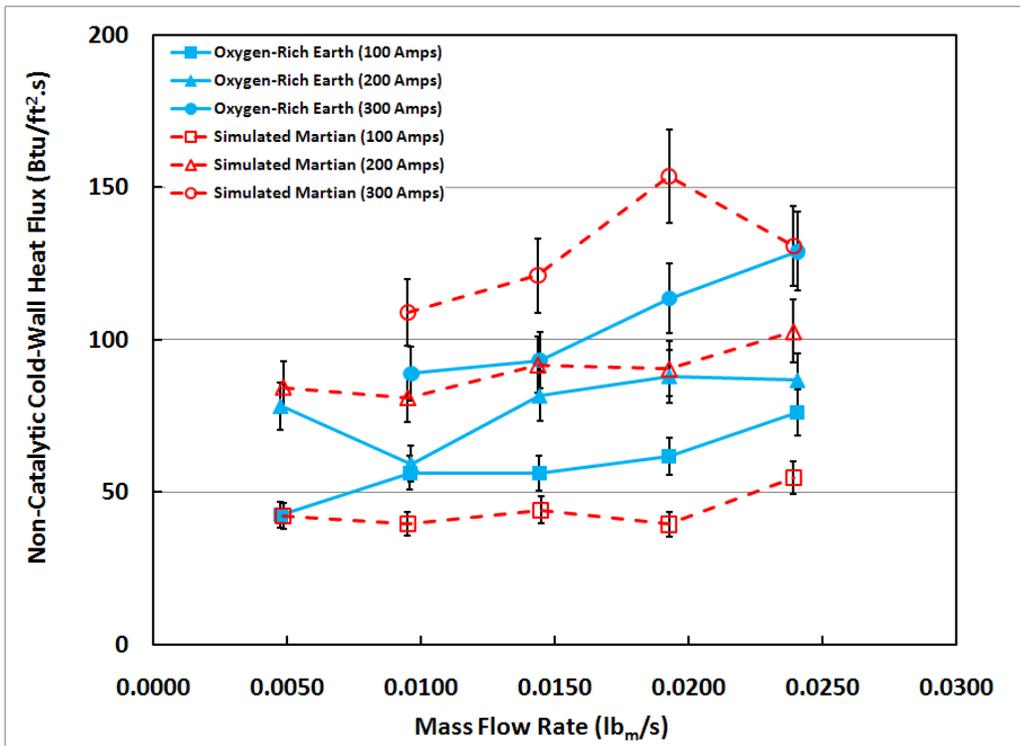


Figure 25. Teflon<sup>®</sup> Slug Calorimeter Cold Wall Heat Flux Comparison

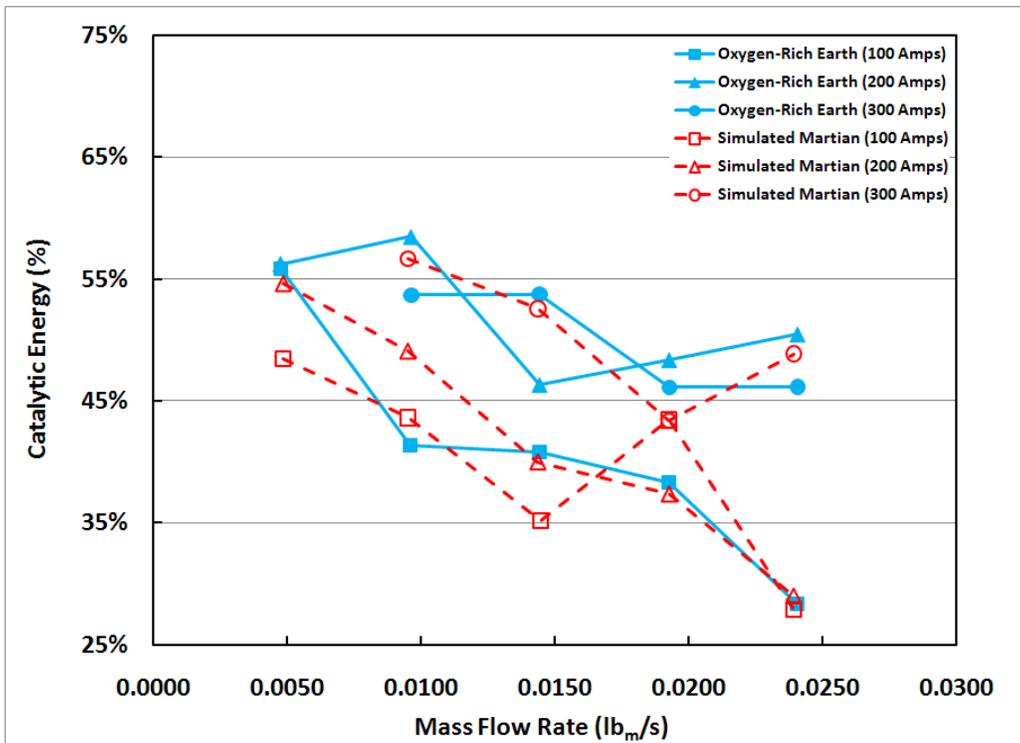


Figure 26. Catalytic Energy in the Plasma Comparison

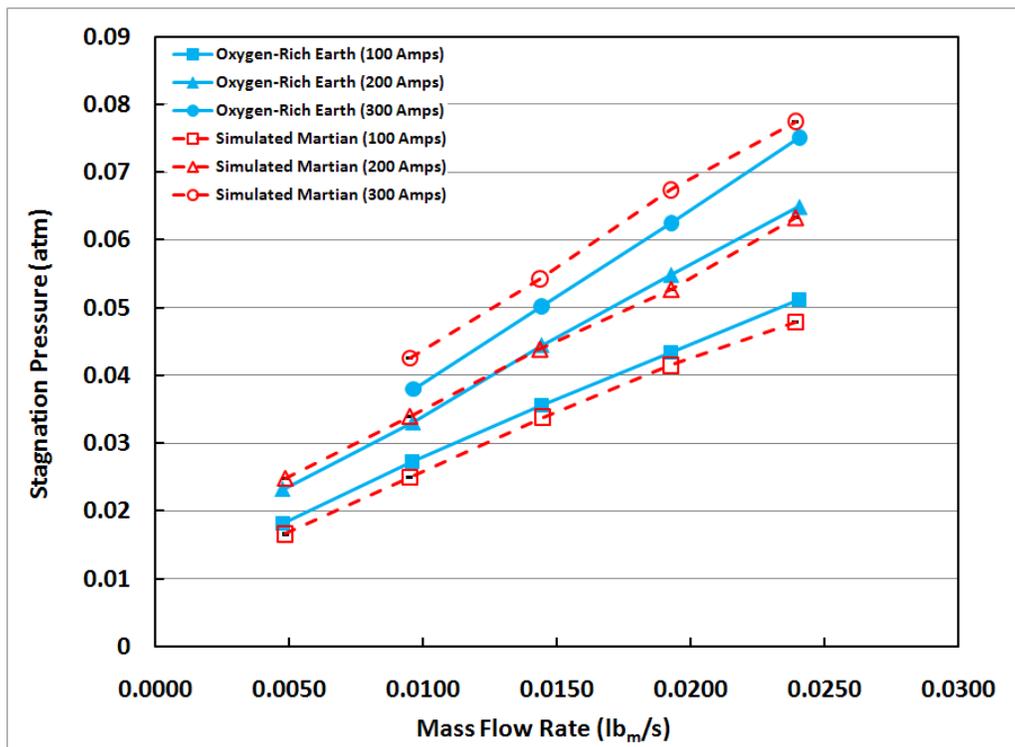


Figure 27. Stagnation Pressure Comparison

Table 7. Oxygen-Rich Earth and Simulated Martian Measurement Differences and Uncertainties

<b>Bulk Enthalpy</b>	<b>RMS (%)</b>	<b>Maximum (%)</b>	<b>Minimum (%)</b>
Oxy. Earth to Sim. Martian Atm. Difference	10.2	13.5	3.4
Oxygen-Rich Earth Atmosphere Uncertainty	7.6	12.0	3.2
Simulated Martian Atmosphere Uncertainty	7.4	11.2	3.1

<b>Fully-Catalytic Cold-Wall Heat Flux</b>	<b>RMS (%)</b>	<b>Maximum (%)</b>	<b>Minimum (%)</b>
Oxy. Earth to Sim. Martian Atm. Difference	22.7	31.0	-30.2
Oxygen-Rich Earth Atmosphere Uncertainty	10.4	10	-10
Simulated Martian Atmosphere Uncertainty	10.4	10	-10

<b>Non-Catalytic Cold-Wall Heat Flux</b>	<b>RMS (%)</b>	<b>Maximum (%)</b>	<b>Minimum (%)</b>
Oxy. Earth to Sim. Martian Atm. Difference	24.8	36.7	-36.0
Oxygen-Rich Earth Atmosphere Uncertainty	10.4	10	-10
Simulated Martian Atmosphere Uncertainty	10.4	10	-10

<b>Stagnation Pressure</b>	<b>RMS (%)</b>	<b>Maximum (%)</b>	<b>Minimum (%)</b>
Oxy. Earth to Sim. Martian Atm. Difference	6.7	12.2	-8.5
Oxygen-Rich Earth Atmosphere Uncertainty	0.21	0.64	0.08
Simulated Martian Atmosphere Uncertainty	0.26	0.81	0.09

### C. Conclusions on Standard Earth, Oxygen-Rich Earth, and Simulated Martian Atmosphere Comparisons

A direct comparison was made between the standard and oxygen-rich Earth atmospheres and the simulated Martian atmosphere which revealed differences in bulk enthalpies, fully-catalytic and non-catalytic heat fluxes, and stagnation pressures for identical facility setpoints. The simulated Martian atmosphere has higher bulk enthalpies than the standard and oxygen-rich Earth atmospheres. The fully-catalytic and non-catalytic cold-wall heat flux measurements for the simulated Martian atmosphere at the 100 Amp current setpoint are lower than the standard and oxygen-rich Earth atmospheres for each of the mass flow rates. At the 200 Amp current setpoint, the heat flux measurements for the simulated Martian atmosphere are higher than the standard and oxygen-rich Earth atmospheres for the lower mass flow rates, but then become lower than the standard and oxygen-rich Earth atmospheres as the mass flow rates increase. Finally, the heat flux measurements for the simulated Martian atmosphere at the 300 Amp current setpoint are higher than the standard and oxygen-rich Earth atmospheres for each of the mass flow rates. The catalytic energy for the oxygen-rich Earth atmosphere compares favorably with that of the simulated Martian atmosphere for each of the facility setpoints. This is in contrast to the higher catalytic energy found in the standard Earth atmosphere. A trend similar to that found in the heat flux measurement comparisons is present in the stagnation pressures for the standard and oxygen-rich Earth atmospheres when compared to the simulated Martian atmosphere at each of the current setpoints.

The oxygen-rich Earth atmosphere and the simulated Martian atmosphere, while containing the same amount of O<sub>2</sub> by weight and being tested at identical facility setpoints, did not have identical results for their bulk enthalpies, fully-catalytic and non-catalytic cold-wall heat flux measurements, and stagnation pressures. The results for the oxygen-rich Earth atmosphere turned out to be a simple offset of the results for the standard Earth atmosphere. This should have been expected as the oxygen-rich Earth atmosphere contains only 4.3% more oxygen by weight than the standard Earth atmosphere. Regardless of the differences, identical test conditions for the oxygen-rich Earth and simulated Martian atmospheres can still be achieved within the envelope of results presented here by manipulating the facility setpoints. This was the first step in comparing an oxygen-rich Earth atmosphere to a simulated Martian atmosphere for material testing. Based on these results, further research is now possible and recommended to compare material performance (i.e. recession rates, char layers, bond-line temperatures, etc.) in the oxygen-rich Earth and simulated Martian atmospheres.

### D. Comparison to European Facilities

Test conditions from an experimental study of several European plasma facilities presented in Refs. 36 and 37 were replicated to within 3% accuracy in HYMETS as shown in Table 8. The fully-catalytic cold-wall heat flux measurements made by the Copper Slug Calorimeter in HYMETS are compared to those made by a Copper Calorimeter Probe in the European facilities L3K, PWK 1 and 2, SIMOUN/COMETE, and the PLASMATRON in Table 9. HYMETS compared favorably to L3K, the only segmented-arc-heated facility in the comparison, at these flow test conditions. The differences between measurements made in HYMETS and L3K were 14% for FTC I and 6% for FTC II. Model geometry, facility configuration, facility power level, and flow property differences can easily account for the variation in the measured heat flux values. Both the COMETE and PLASMATRON facilities produce subsonic plasma flows which provide lower heat flux levels when directly compared to the supersonic plasma flows from the segmented-arc-heated facilities. An indirect comparison can be made by normalizing both subsonic and supersonic heat flux values by the method demonstrated in Ref. 37. The normalized comparison establishes that L3K and the PLASMATRON agree to within 3% of each other for these test conditions<sup>37</sup>, indicating that HYMETS also compares favorably to the PLASMATRON facility by extension. Without normalization, the results for the PWK 1 and 2, and the COMETE facilities do not compare favorably with HYMETS. Normalization of the PWK 1 and 2, and the COMETE facilities was not performed in Ref 37.

**Table 8. HYMETS / European Flow Test Conditions (FTC)**

<b>FTC-I</b>	<b>HYMETS</b>	<b>European</b>	<b>Difference (%)</b>
Stag. Press. (atm)	0.0355	0.0345	2.90
Bulk Enth. (Btu/lb <sub>m</sub> )	3895	3870	0.65

<b>FTC-II</b>	<b>HYMETS</b>	<b>European</b>	<b>Difference (%)</b>
Stag. Press. (atm)	0.0349	0.0345	1.16
Bulk Enth. (Btu/lb <sub>m</sub> )	5596	5589	0.11

**Table 9. Fully-Catalytic Cold-Wall Heat Flux (Btu/ft<sup>2</sup>.s)**

Test Conditions	HYMETS	European Plasma Facilities*			
		L3K	PWK 1 & 2	SIMOUN/ COMETE	PLASMATRON
FTC-I	105	92	---	---	39
FTC-II	126	119	202	32	70

\*described in Ref. 36

## V. Summary

Future robotic and manned missions to Mars with a significant increase in landed mass and a corresponding increase in aerothermodynamic heating will require the recertification of heritage TPS or the certification of new TPS in relevant environments. The test capability to certify TPS in a relevant CO<sub>2</sub> environment to simulate Martian entry disappeared in the United States after the early 1970's. To compensate for the lack of testing in CO<sub>2</sub>, response models were indirectly validated by varying the percentage of oxygen in arc-jet tests. The difference between tests in air and CO<sub>2</sub> is the amount of available oxygen, and it is unknown whether an oxygen-rich environment is comparable to a CO<sub>2</sub> environment. NASA Langley Research Center's HYMETS facility was recently modified to provide a CO<sub>2</sub> test environment. An overview of the modification, operation, and performance of the HYMETS arc-jet facility with CO<sub>2</sub> as a test gas was provided. A range of test conditions, or envelope map, for the HYMETS facility was performed with a standard and oxygen-rich Earth atmosphere and a simulated Martian atmosphere. Of particular interest is a comparison of results between the oxygen-rich Earth atmosphere and the simulated Martian atmosphere to assess the adequacy of testing in an oxygen-rich Earth atmosphere as a replacement for testing with CO<sub>2</sub>. Additional tests were performed in HYMETS with test conditions replicating those from an experimental study of several European plasma facilities.

The current operational CO<sub>2</sub> gas ratio for HYMETS is 24% N<sub>2</sub>, 71% CO<sub>2</sub>, and 5% Ar by volume. This provides a 25% to 75% ratio by volume of N<sub>2</sub> to CO<sub>2</sub> when Ar is excluded from the mix. This current ratio is closer to achieving the actual gas ratio for the Martian atmosphere of 3% N<sub>2</sub> to 97% CO<sub>2</sub> than historical attempts. For reference, the operational air ratio for HYMETS is 75% N<sub>2</sub>, 20% O<sub>2</sub>, and 5% Ar by volume. This provides a 79% to 21% ratio by volume of N<sub>2</sub> to O<sub>2</sub> when Ar is excluded from the mix, which is the actual gas ratio for the Earth atmosphere. Modifications to the facility included a tube and shell heat exchanger with three pressure-regulated stages for the CO<sub>2</sub> gas distribution lines, and eight supply cylinders were drawn from simultaneously to prevent the formation of dry-ice during sustained facility operation. Additionally, CO<sub>2</sub> introduced a "soot-like" deposit onto the components of the arc plasma generator which decreased the stability of the electric arc and required regular cleaning. A potentially explosive mix of CO and O<sub>2</sub> was produced during operation of the facility with CO<sub>2</sub>. However, the explosive mix was not combustible below the maximum operating pressure for the facility of 25.4 torr, and the facility was purged with N<sub>2</sub> and Ar before and after operation with CO<sub>2</sub>. The production of particulate CN or HCN during facility operation with CO<sub>2</sub> and N<sub>2</sub> was determined to be negligible. The production of NO during facility operation with CO<sub>2</sub> and N<sub>2</sub> was confirmed by RGA measurements and NO PLIF measurements, however, the amounts were within the existing limits for the HYMETS air permit. Higher ratios of CO<sub>2</sub> to N<sub>2</sub> than the current ratio, 75% CO<sub>2</sub> to 25% N<sub>2</sub> by volume, resulted in unacceptable oxidation of the cathode. A possible method for future work in achieving higher ratios is to introduce CO near the cathode to reduce the amount of N<sub>2</sub> needed to shield the cathode. The tungsten button in the cathode is not easily oxidized by CO.

A direct comparison was made between the standard and oxygen-rich Earth atmospheres and the simulated Martian atmosphere which revealed differences in bulk enthalpies, fully-catalytic and non-catalytic heat fluxes, and stagnation pressures for identical facility setpoints. The simulated Martian atmosphere has higher bulk enthalpies than the standard and oxygen-rich Earth atmospheres. The fully-catalytic and non-catalytic cold-wall heat flux measurements for the simulated Martian atmosphere at the 100 Amp current setpoint are lower than the standard and oxygen-rich Earth atmospheres for each of the mass flow rates. At the 200 Amp current setpoint, the heat flux measurements for the simulated Martian atmosphere are higher than the standard and oxygen-rich Earth atmospheres for the lower mass flow rates, but then become lower than the standard and oxygen-rich Earth atmospheres as the mass flow rates increase. Finally, the heat flux measurements for the simulated Martian atmosphere at the 300 Amp current setpoint are higher than the standard and oxygen-rich Earth atmospheres for each of the mass flow rates. The catalytic energy for the oxygen-rich Earth atmosphere compares favorably with that of the simulated Martian atmosphere for each of the facility setpoints. This is in contrast to the higher catalytic

energy found in the standard Earth atmosphere. A trend similar to that found in the heat flux measurement comparisons is present in the stagnation pressures for the standard and oxygen-rich Earth atmospheres when compared to the simulated Martian atmosphere at each of the current setpoints.

The oxygen-rich Earth atmosphere and the simulated Martian atmosphere, while containing the same amount of O<sub>2</sub> by weight and being tested at identical facility setpoints, did not have identical results for their bulk enthalpies, fully-catalytic and non-catalytic cold-wall heat flux measurements, and stagnation pressures. The results for the oxygen-rich Earth atmosphere turned out to be a simple offset of the results for the standard Earth atmosphere. This should have been expected as the oxygen-rich Earth atmosphere contains only 4.3% more oxygen by weight than the standard Earth atmosphere. Regardless of the differences, identical test conditions for the oxygen-rich Earth and simulated Martian atmospheres can still be achieved within the envelope of results presented here by manipulating the facility setpoints. This was the first step in comparing an oxygen-rich Earth atmosphere to a simulated Martian atmosphere for material testing. Based on these results, further research is now possible and recommended to compare material performance (i.e. recession rates, char layers, bond-line temperatures, etc.) in the oxygen-rich Earth and simulated Martian atmospheres.

Finally, two different test conditions from an experimental study of several European plasma facilities were replicated to within 3% accuracy. HYMETS compared favorably to European plasma facility L3K, the only segmented arc heated facility in the comparison, with differences in fully-catalytic cold-wall heat flux of 14% and 6% for the two test conditions. A normalized comparison established that L3K and the PLASMATRON, another European plasma facility, agreed to within 3% of each other for these test conditions, indicating that HYMETS also compared favorably to the PLASMATRON facility by extension.

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