



## Comparing On-Orbit and Ground Performance for an S-Band Software-Defined Radio

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- Introduction to Software Defined Radio (SDR)
  - Why SDR?
  - Space Communications and Navigation (SCaN) Testbed
- Pre-launch Characterization
- Design of a Received Power Estimator
  - Ground development
  - Space performance



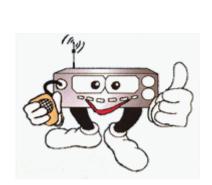


Software-defined radio (SDR) – a modern communication platform

FLEXIBLE!

- Radio frequency module
- Signal processing module [waveform]
- General processing module
- SDR is...

# ADAPTABLE!



#### HourTTac HourTT

**General Processing Module (GPI** 

General Purpose Proces

Ground Test Interface

# PREDICTABLE ...?

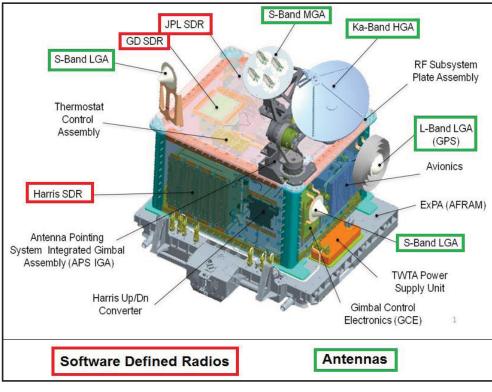




- Space Communications and Navigation (SCaN) Testbed
  - External payload on the International Space Station (ELC-3 location)



SCaN Testbed installed to the ExPRESS Logistics Carrier-3

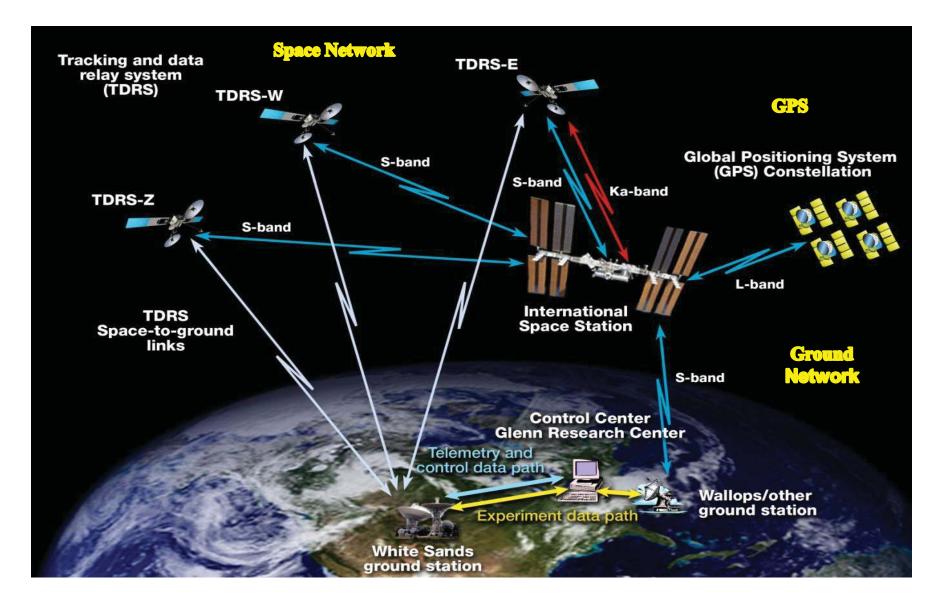


SCaN Testbed hardware block diagram



# **STB Experiment Communication**









- Jet Propulsion Laboratory (JPL) SDR part of STB
  - S-band transceiver (7 Watts) with L-band receive capability
  - 66 MHz SPARC (RTEMS) processor and 2 Virtex-II FPGAs

Int'l Space Station

- Three JPL SDRs!
  - Flight model (FM)
    - Radio Frequency Module, Global Positioning System Module, Baseband Processing Module, Power Amplifier / Power Supply Module

Ground

- Engineering model (EM)
  - Same as FM, except commercial grade parts.



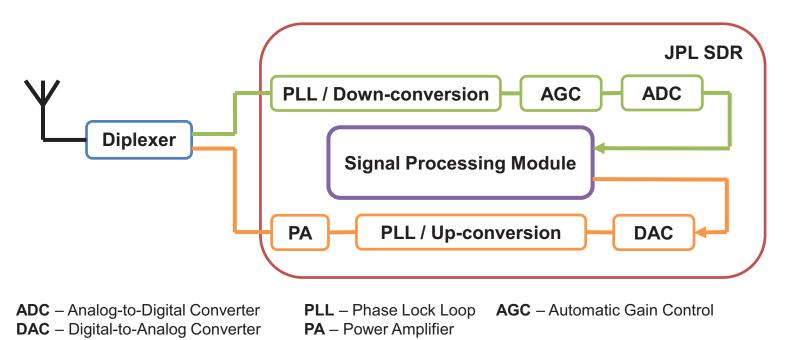
- Breadboard
  - Baseband Processing Module only.







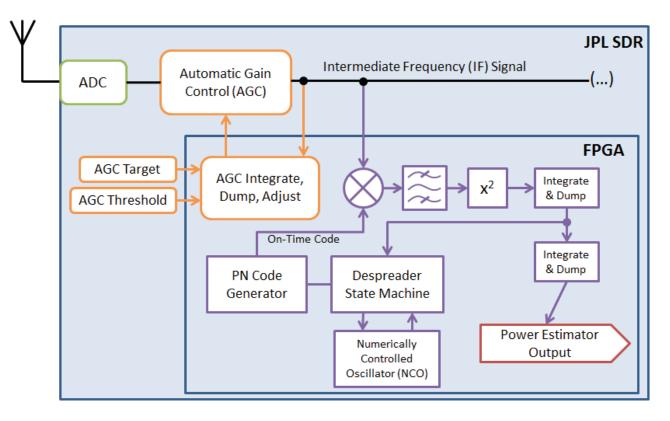
- Flight model SDR testing prior to launch
  - Establish a performance baseline in a controlled environment
  - Collect data useful for future waveform capabilities
- Lesson Learned test the hardware independent of the waveform
  - Test very close to hardware interfaces
  - Do not make testing dependent on software implementation







- Estimating received power is a useful diagnostic feature
- Uses existing waveform despreader digital filters
  - Performed at the intermediate frequency (IF) after downconversion
  - BPSK filter bandwidth = 2\*(signal bandwidth) + (Doppler allowance)
  - Despreader PN generator is bypassed for non-spread modes.



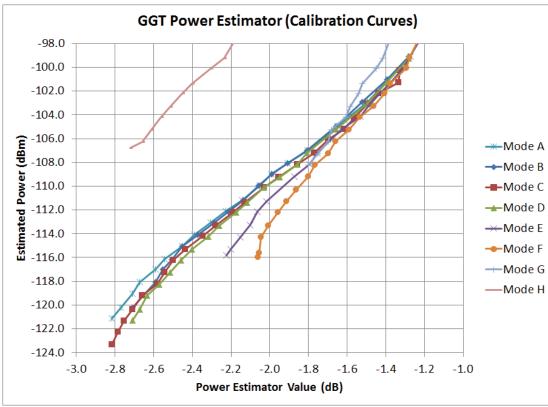




### • Performed testing on the engineering model

- Map the "Integrate & Dump" value to the corresponding input power
- Swept input power level across realistic space received power range
- Power Estimate = Signal Power + Noise Power
- Waveform "mode" → data rate, frequency, spreading, etc.

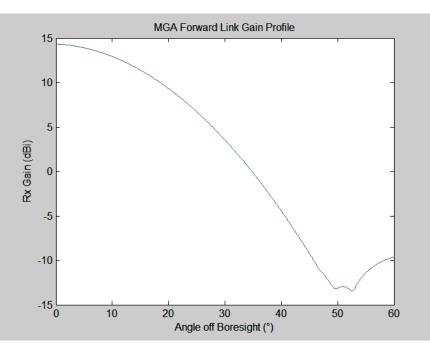
Mode	Spread	Symbol Rate (ksps)	Freq. MHz	Filter BW (kHZ)	
Α	Yes	18	2106	149	
В	Yes	36	2106	188	
С	Yes	18	2041	149	
D	Yes	36	2041	188	
E	No	155	2041	450	
F	No	310	2041	789	
G	No	769	2041	1793	
Н	No	1538	2041	3468	

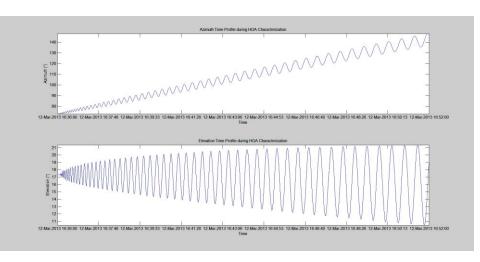






- Limited power range and test time in space
  - Space link varies by ~2 dB due to distance over ~40 minutes
  - NASA satellites have 2 fixed transmit power levels
- Implemented spiral motion on the MGA
  - Swept elevation over a wide range of power (~20 dB) during 1 pass
  - Used 1-degree lap size based on in-situ antenna pattern

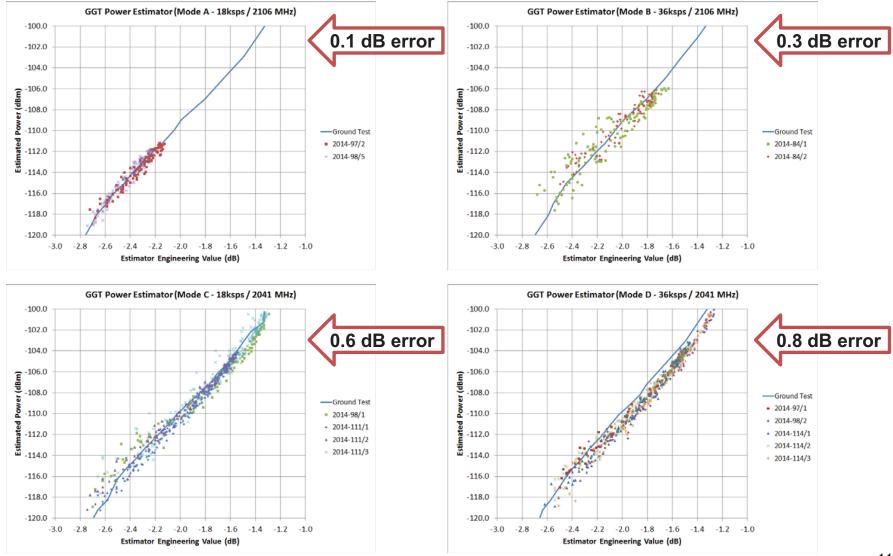






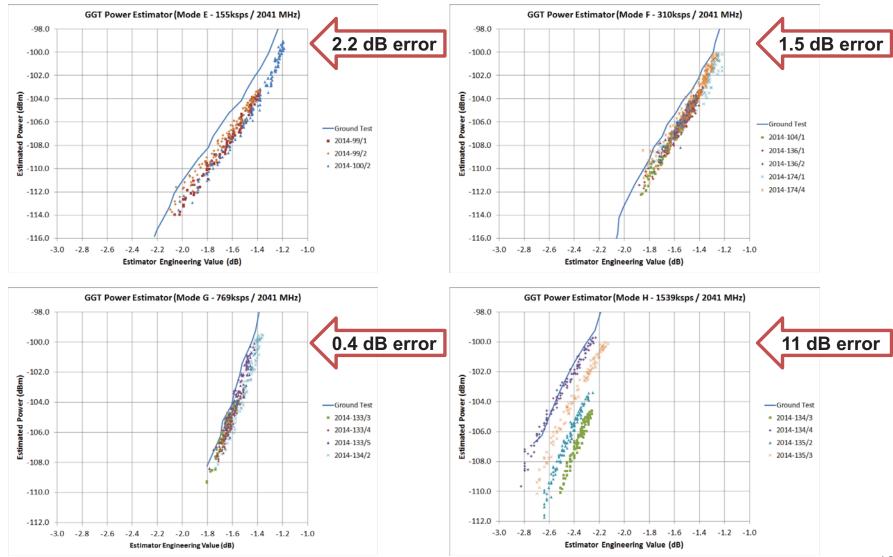


#### • Spread-spectrum results versus engineering model performance













- Overall the power estimator performance is acceptable.
  - Spread waveform modes show less than 1 dB average error
  - Non-spread modes show 1 to 2 dB average error (except mode H)
- The power estimator is sensitive to AGC fluctuation.
  - AGC level directly affects the IF power level
  - Mode H has a very low AGC set point  $\rightarrow$  11 dB average error!

## Future work

- Improve understanding of how wideband noise affects the AGC algorithm
- Incorporate AGC level into the power estimator
- Look into narrower filter bandwidths for lower received power levels





