



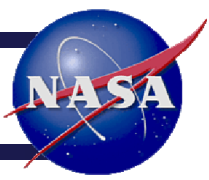
GSFC · 2015

Infrared Images of Boundary Layer Transition on the D8 Transport Configuration in the LaRC 14- by 22-Foot Subsonic Tunnel

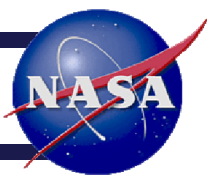
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Virginia 23681*



Presentation Outline



Topic	Slides
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• Historical Perspective	4, 5
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The purpose of this study is to non-intrusively detect the location of laminar to turbulent boundary layer transition on a D8 transport model in the LaRC 14'x22' Subsonic Tunnel.

Goal

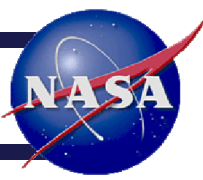
- Determine the effectiveness of trip tape applied around the nose of the model, and along the leading edge of the wings, for different flow conditions

Study Impact

- Infrared thermography provides a non-intrusive technique to compare the expected and observed boundary layer transition locations for the given flow conditions and implemented boundary layer trip mechanism
- In subsonic flows, the surface of the model quickly reaches an equilibrium temperature in regions with both laminar and turbulent boundary layers, so infrared thermography images must be recorded at the correct time to observe transition
- Infrared radiation from the model also can demonstrate other flow phenomena over the surface of the wind tunnel model in addition to boundary layer transition

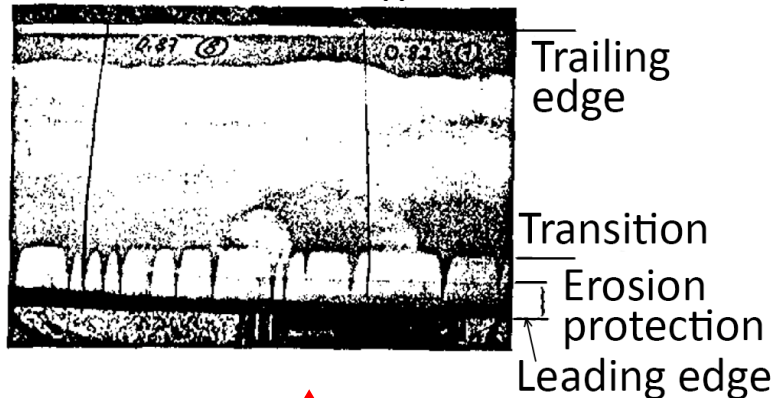


Historical Perspective – Sublimation Techniques

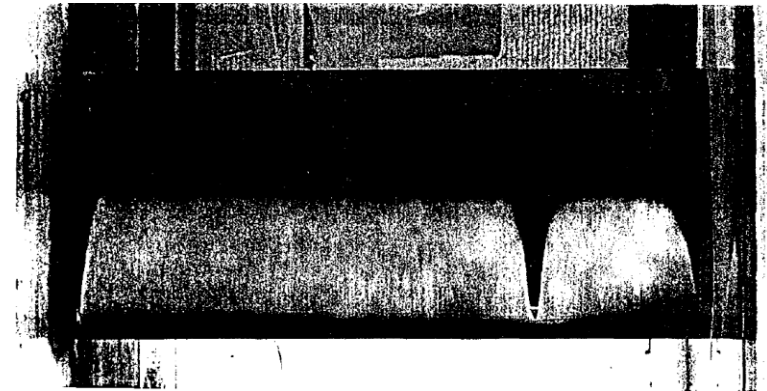


Acenaphthene on a helicopter rotor blade
(Körner et al., AIAA 87-0085)

Upper side $0.82 \leq \frac{r}{R} \leq 0.87$



Napthalene on a wooden model at low angle of attack
(Crowder, AIAA 90-1450)



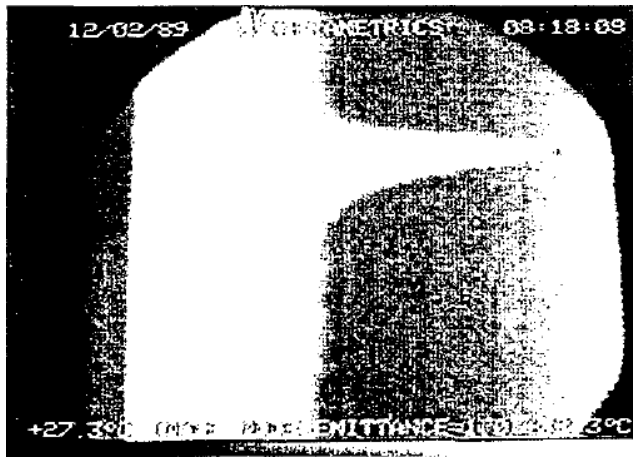
Boundary layer transition studies used to require a coating of a sublimating material, such as acenaphthene or naphthalene, on the surface of models or flight vehicles. This sublimating technique only yields one transition image per application of the coating.



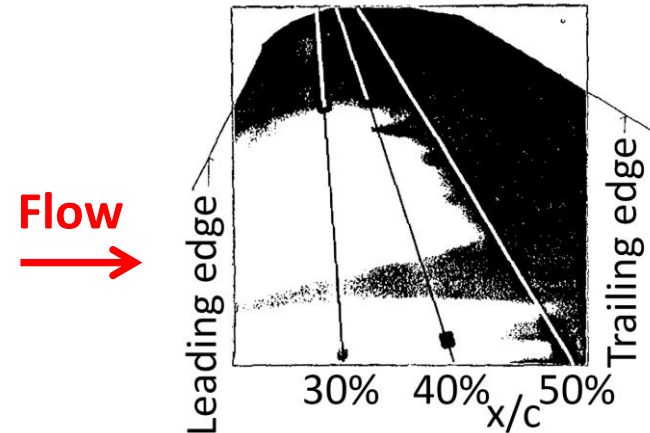
Historical Perspective – Infrared Radiation (IR)



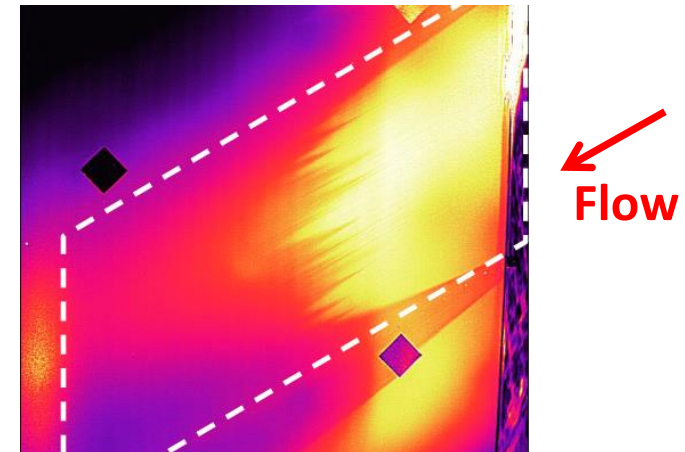
IR image of transition on a wooden model at low angle of attack (Crowder, AIAA 90-1450)



IR visualization of transition in flight (Körner et al., AIAA 87-0085)



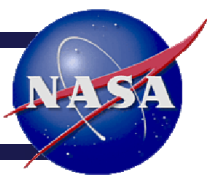
ExamInIR processed IR image of an airfoil (Crawford et al., AIAA 2014-1411)



The current capability of IR imaging for the purpose of transition studies is vastly improved in terms of temperature and spatial resolution.



Experimental Facility

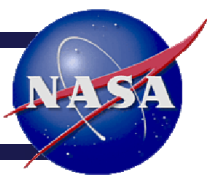


14- by 22-Foot Subsonic Tunnel





MIT D8 Model Installed in Test Section

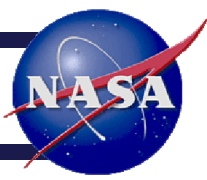


August 4, 2015

TFAWS 2015 – August 3-7, 2015 – Silver Spring, MD

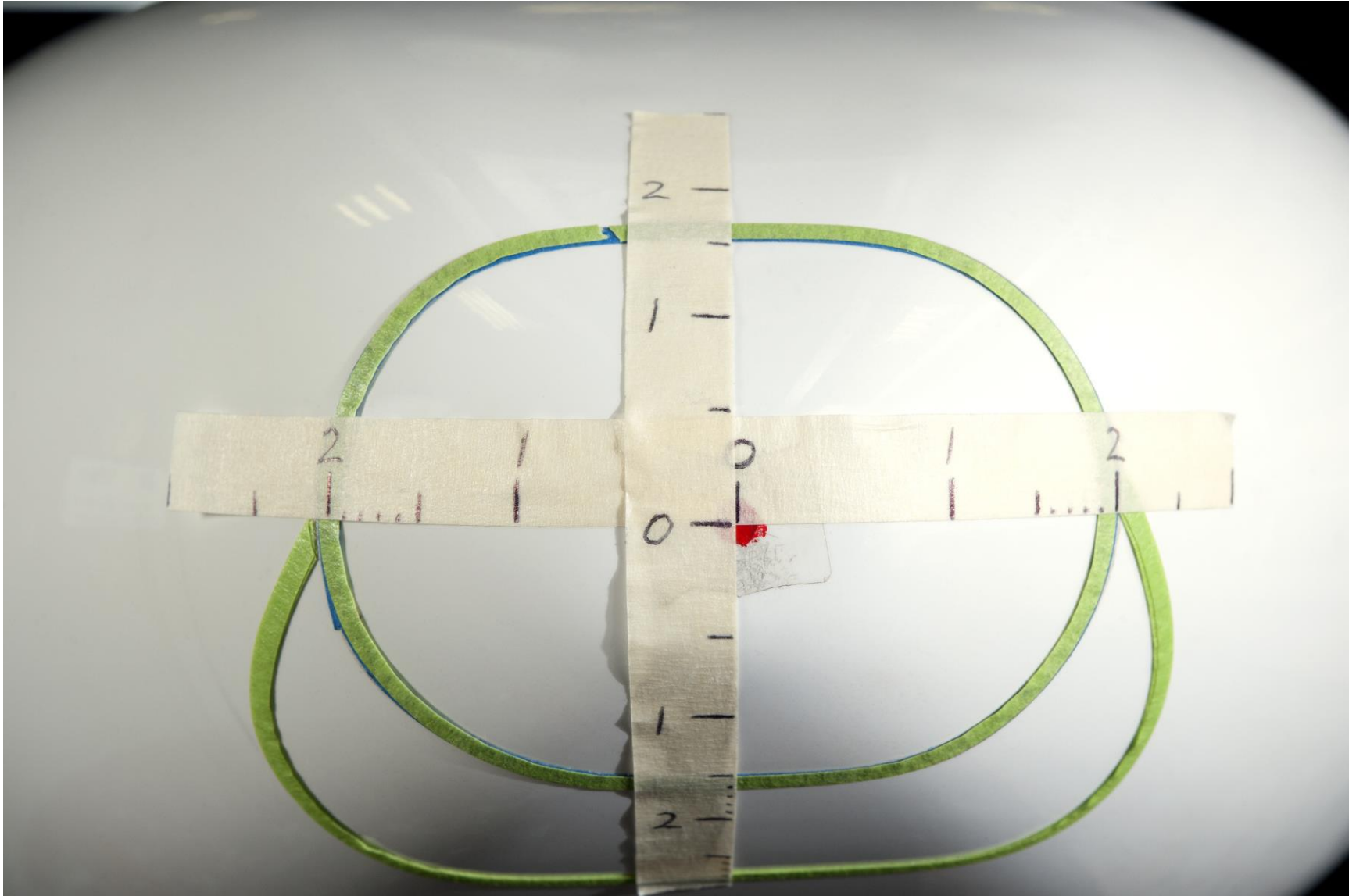


MIT D8 Model Installed in Test Section



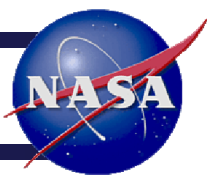


Trip Tape at Fuselage Nose





IR Camera Mounted in Side Wall



Long wavelength FLIR SC 3000 IR Camera

Spectral responsivity: 8-9 μ m

Temperature resolution of 20mK at 30°C

Spatial resolution of 320x240 pixels



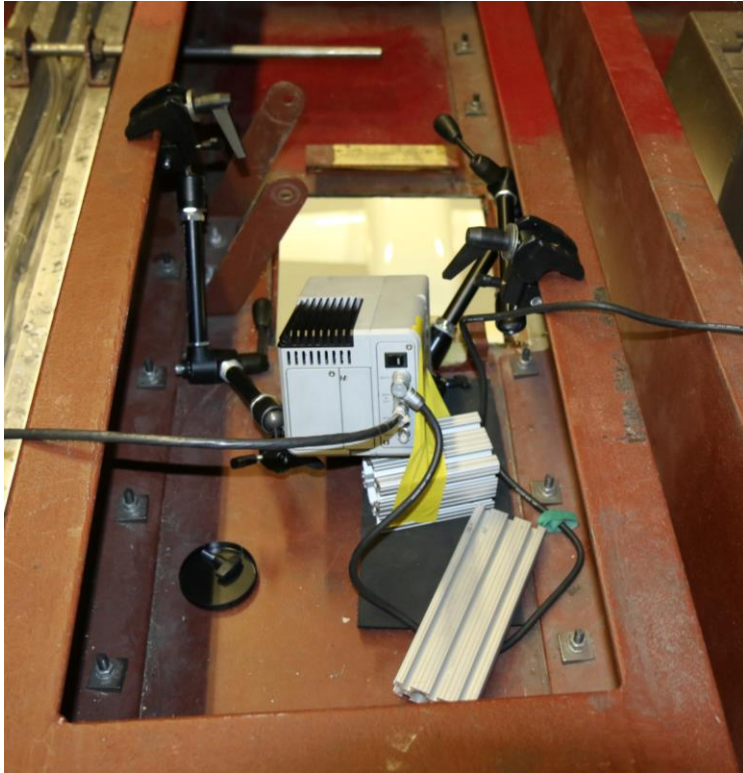
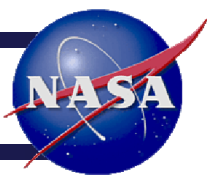
Outside the tunnel, camera aimed at the model from the side wall



Inside the tunnel, metal blank with hole replaced a pane of glass in tunnel side wall



IR Camera Mounted on top of Ceiling



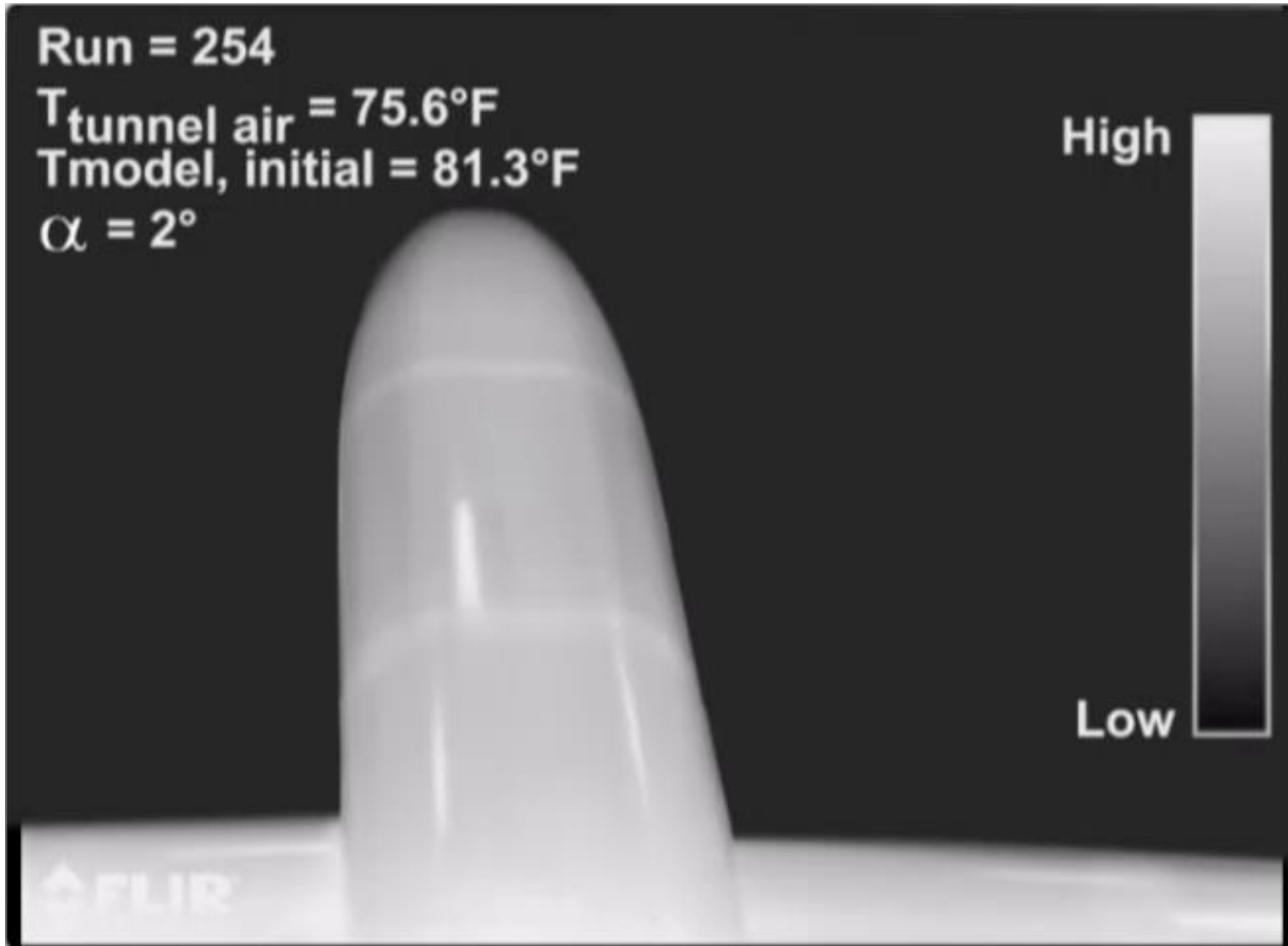
Top view, camera aimed at the model from the ceiling



Bottom view, showing location of removed pane of glass in tunnel ceiling

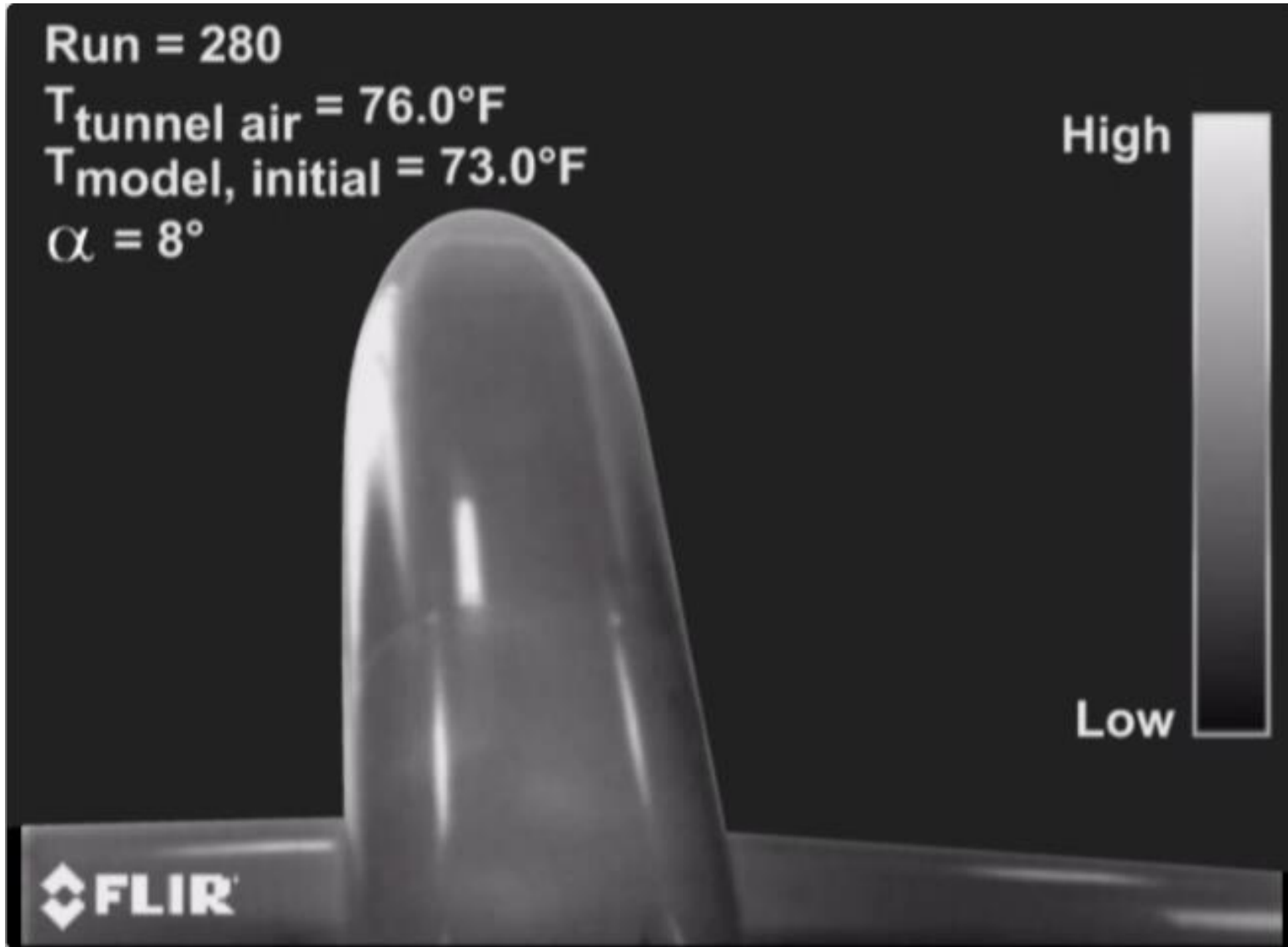
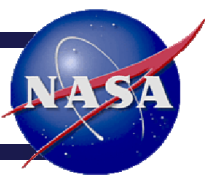


Turbulent regions cooler than laminar regions



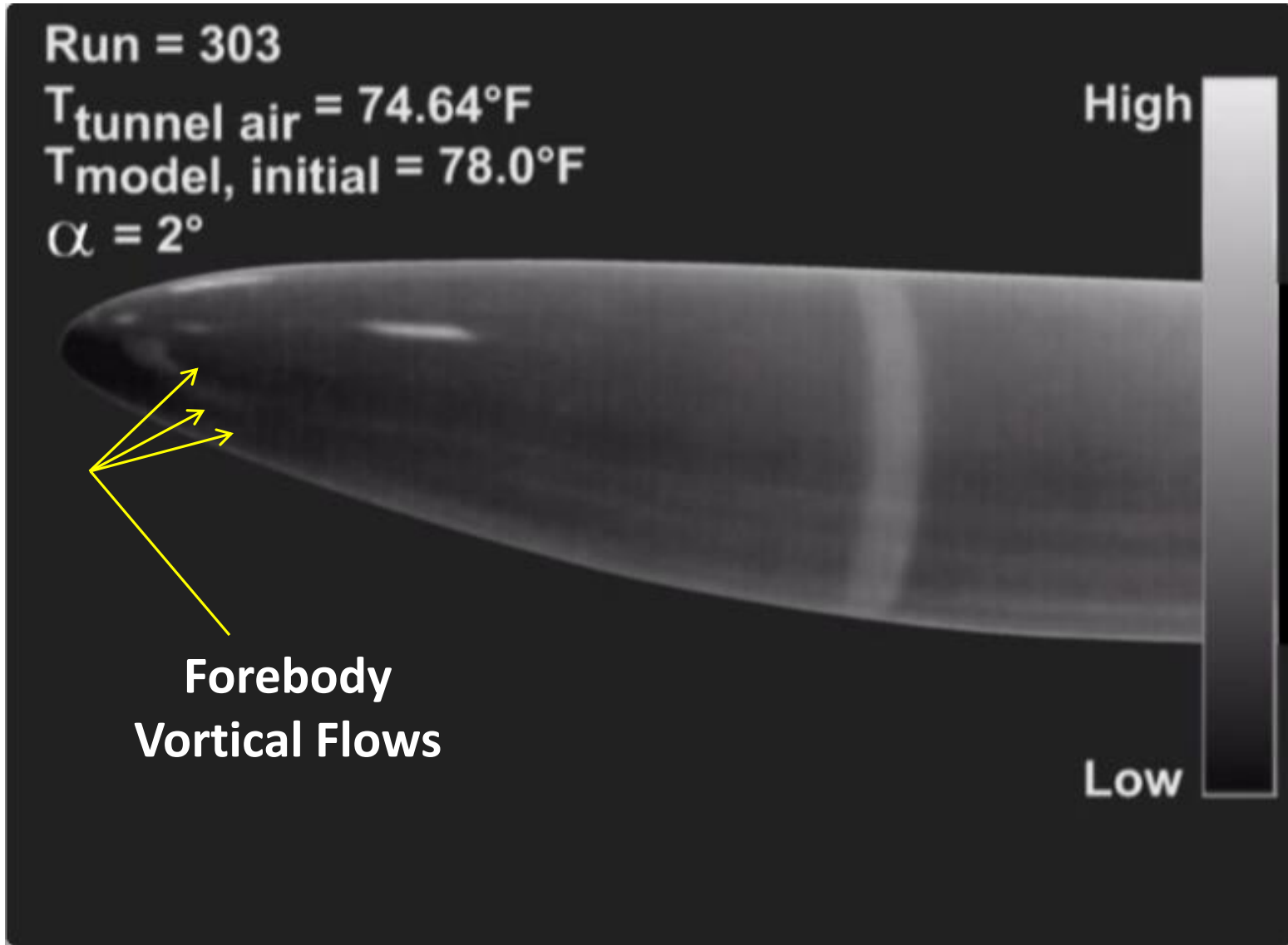
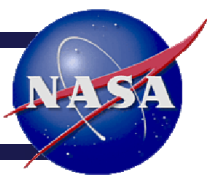


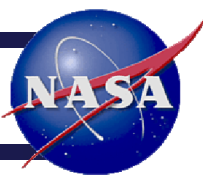
Turbulent regions warmer than laminar regions



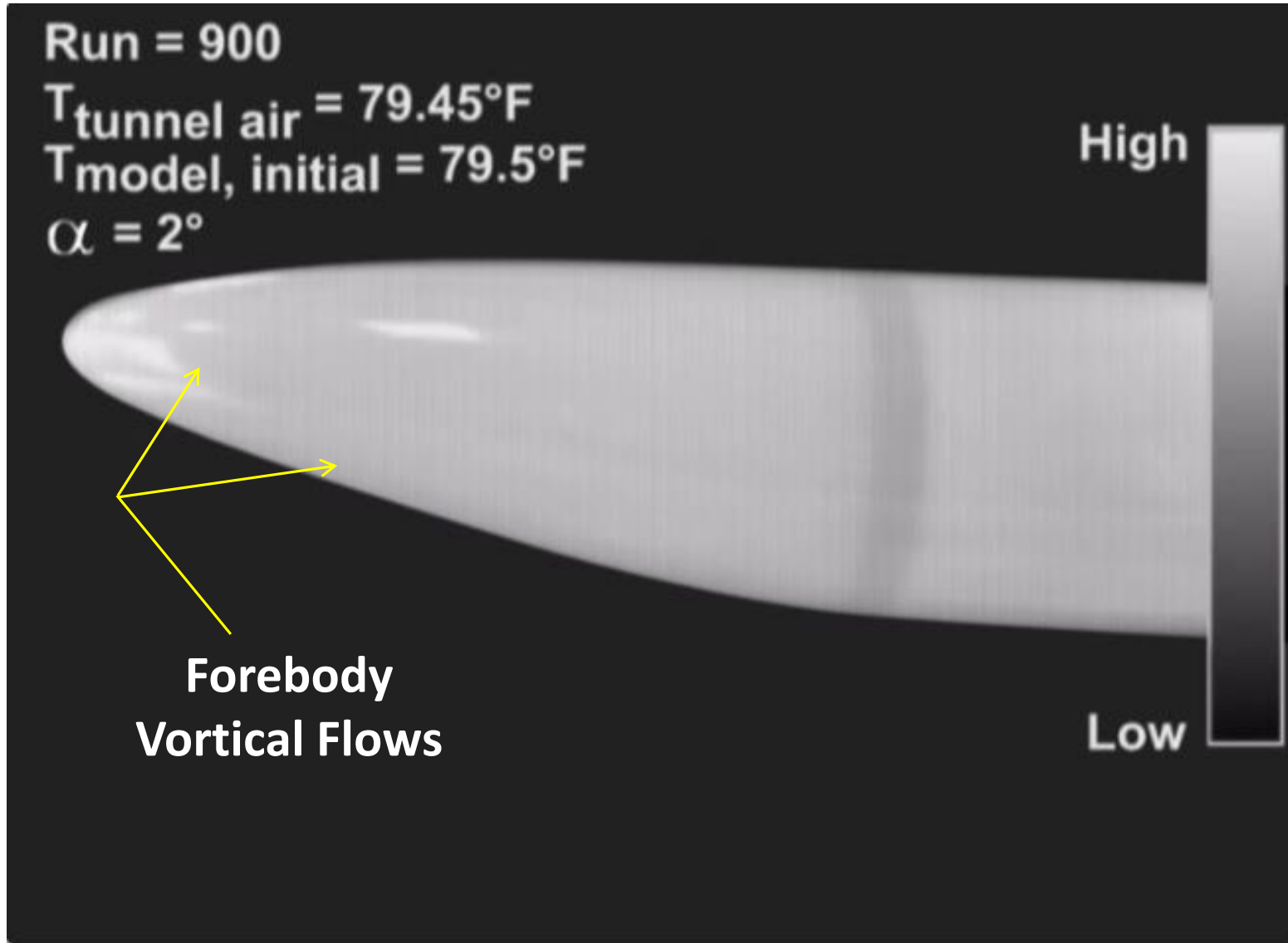


Turbulent regions cooler than laminar regions



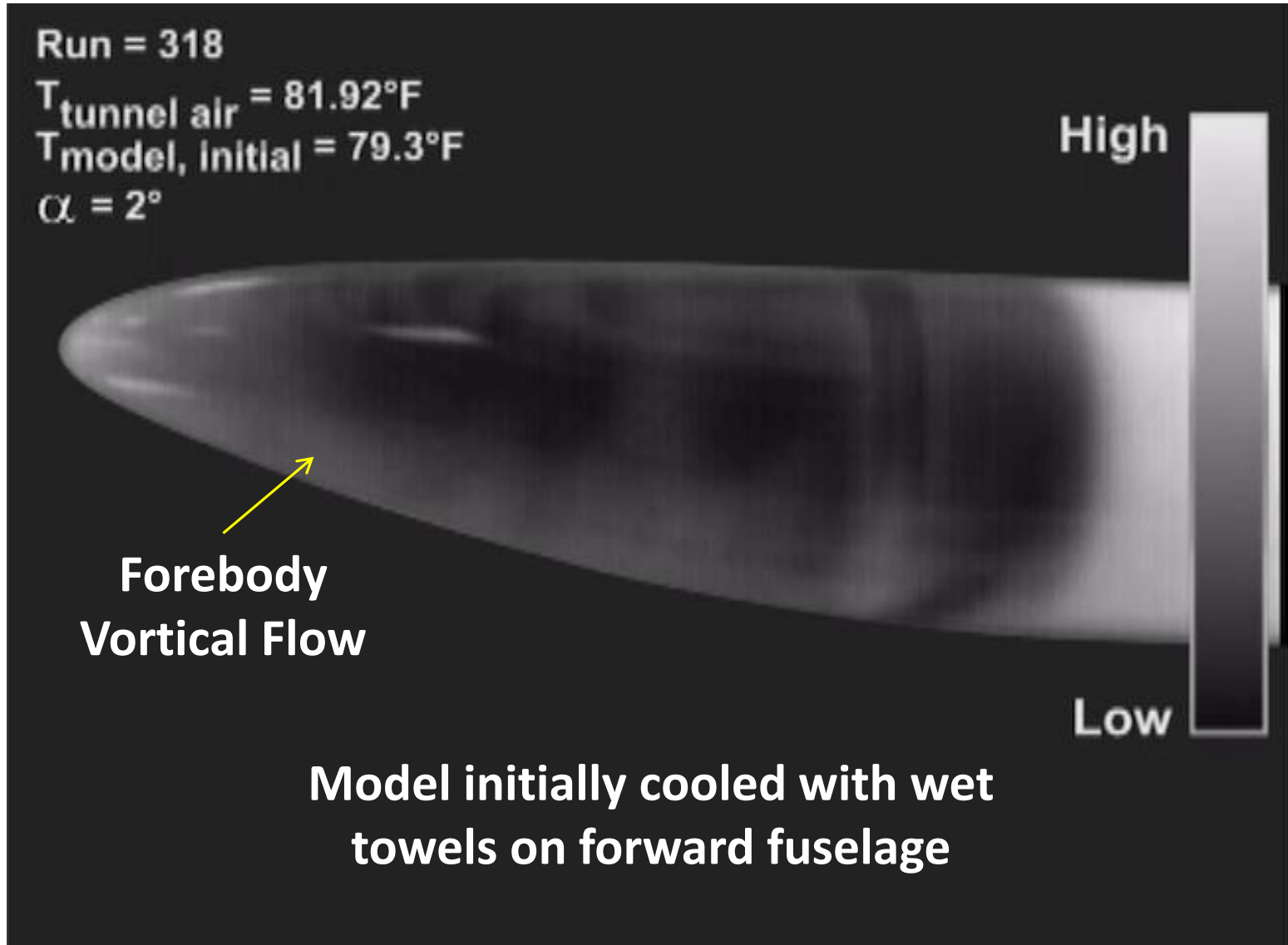
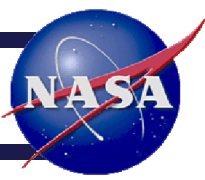


Turbulent regions warmer than laminar regions



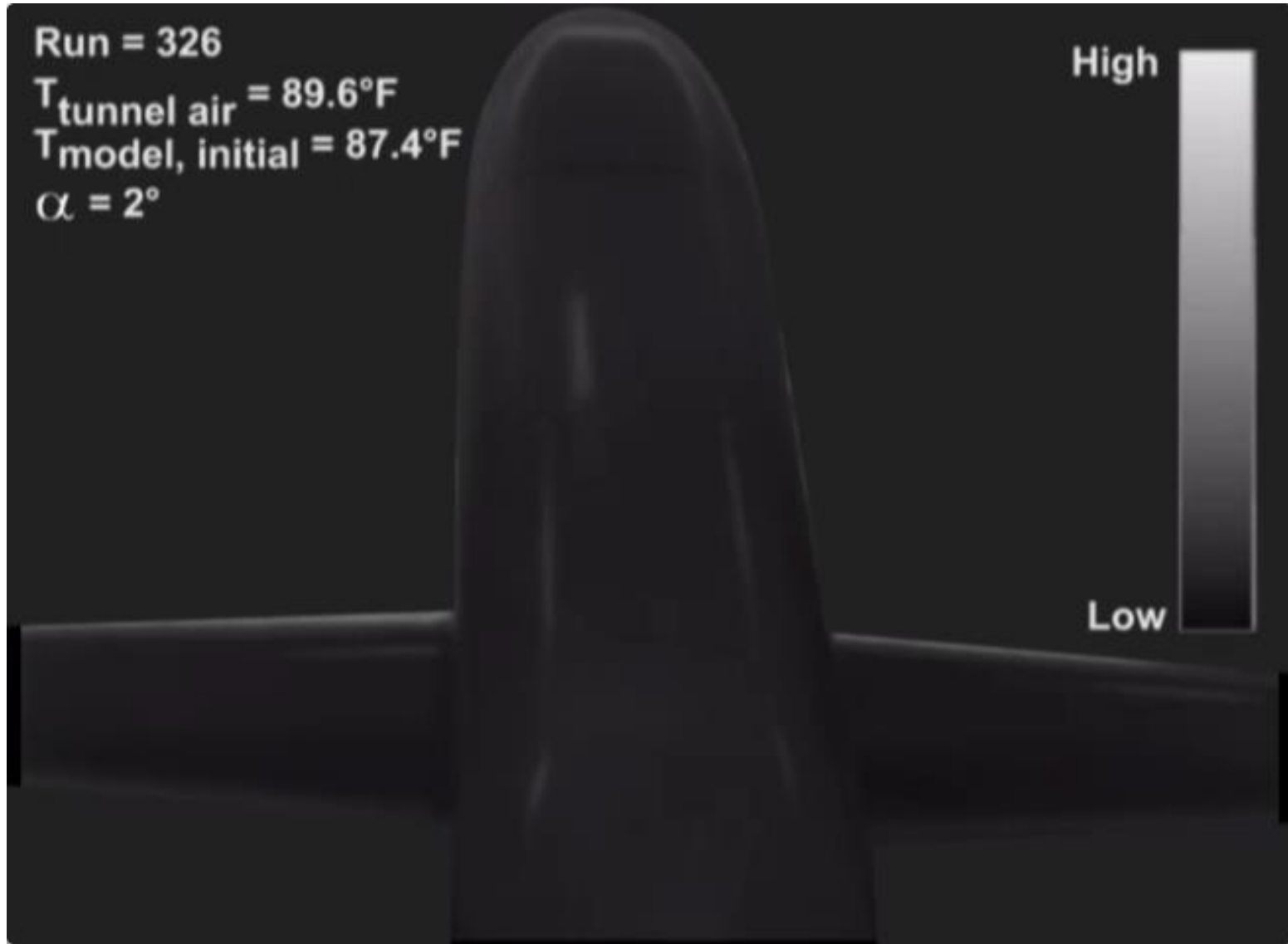


Turbulent regions warmer than laminar regions



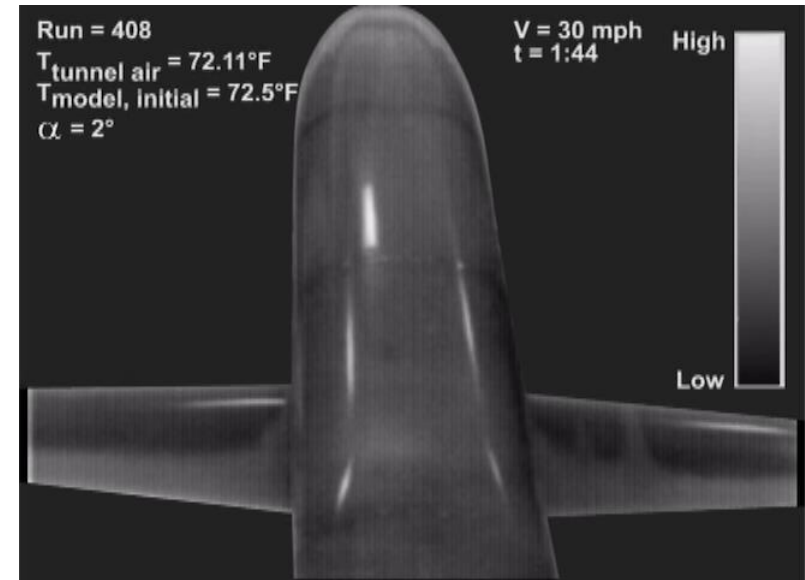
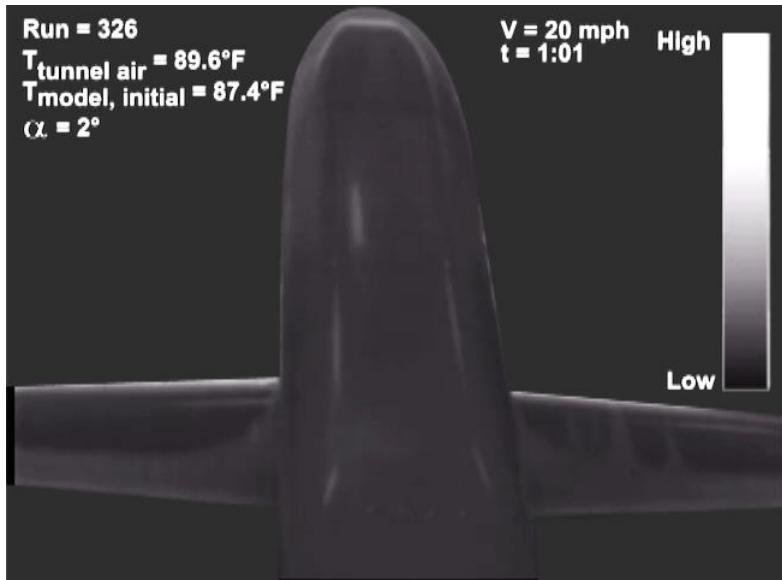
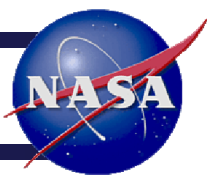


Natural and forced transition on wings

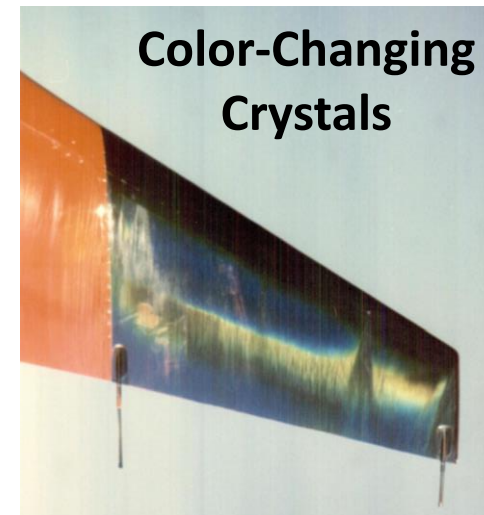




Natural and forced transition on wings

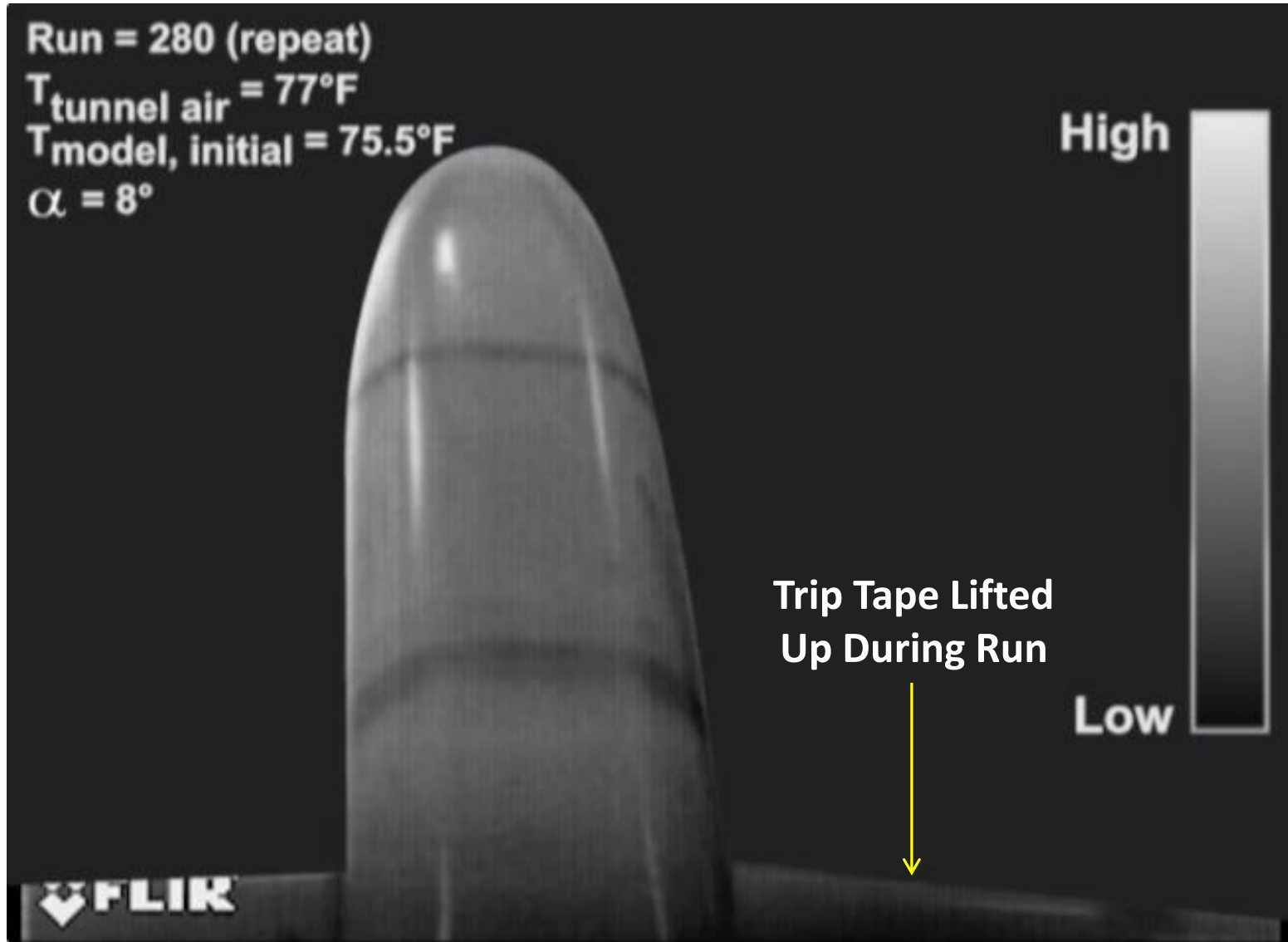
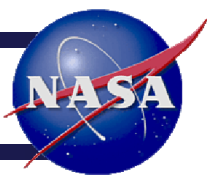


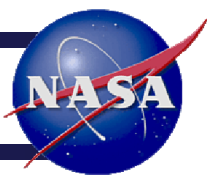
IR cameras provide the capability to non-intrusively observe natural and forced transition that previously was measured by more elaborate techniques such as coating the vehicle or model with either color-changing crystals or a sublimating material.



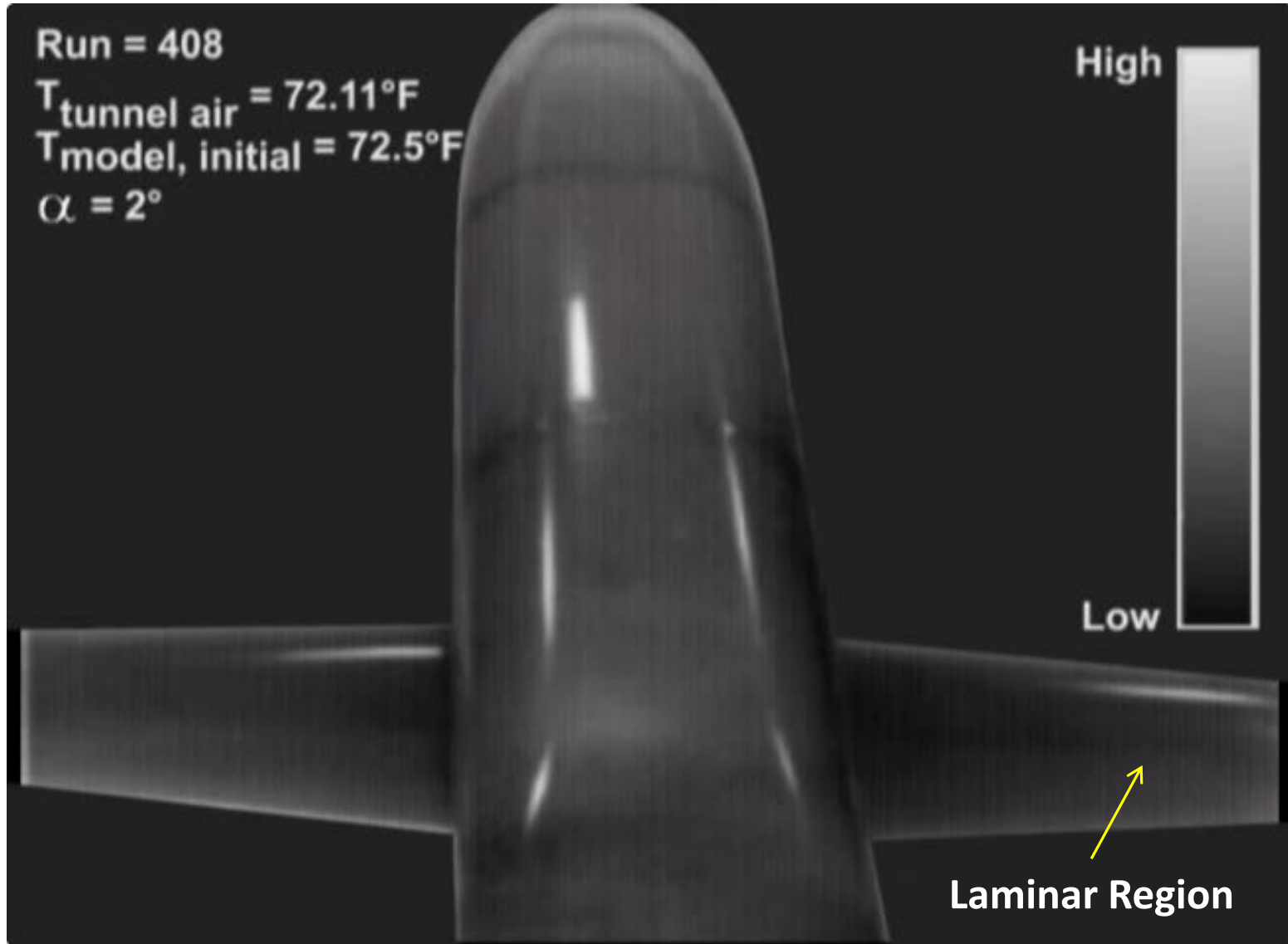


Trip tape problem identified



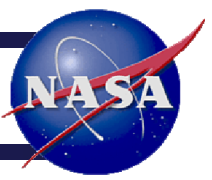


Laminar region behind removed trip tape





Turbulent regions warmer than laminar regions

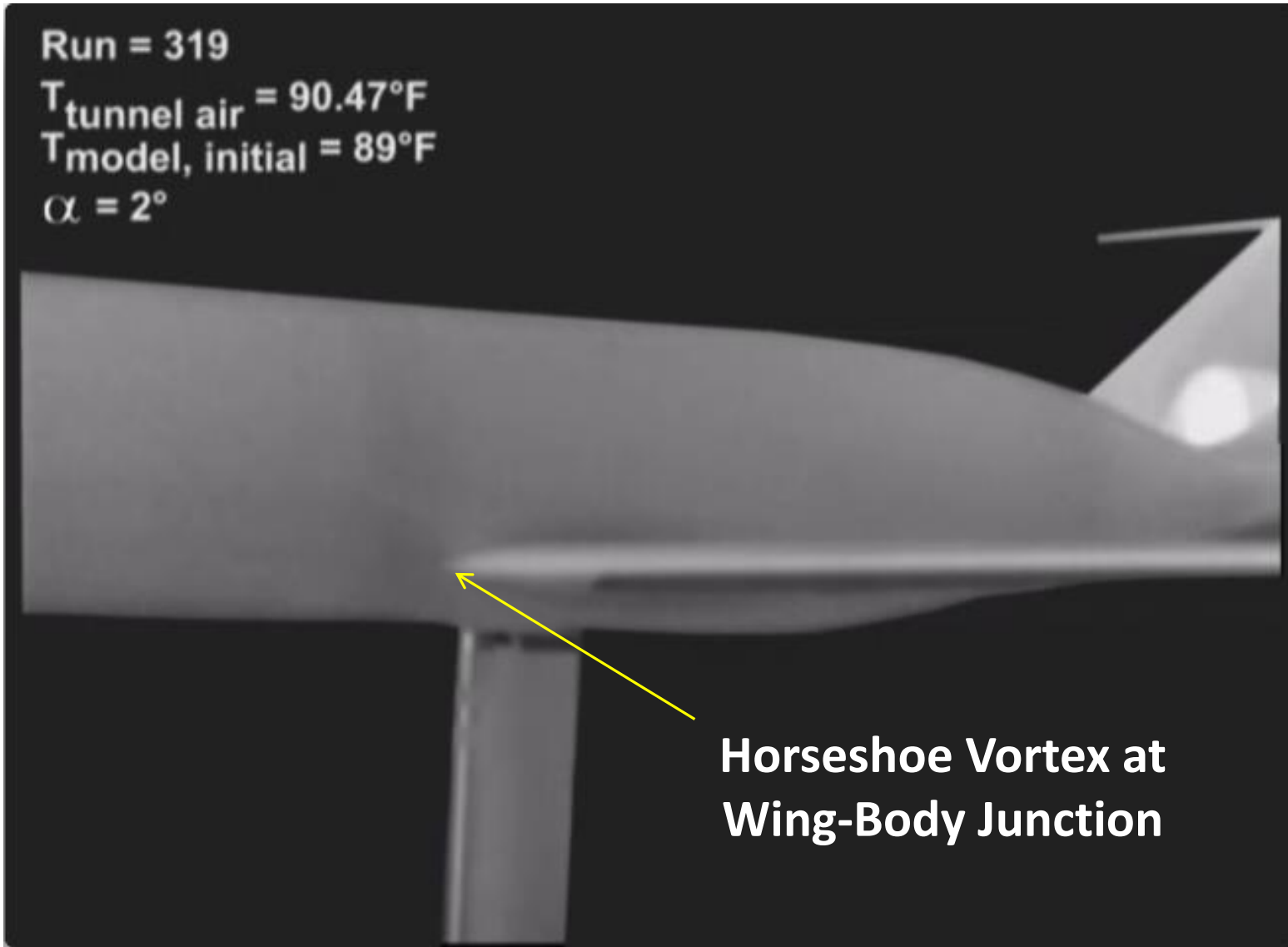


Run = 319

$T_{\text{tunnel air}} = 90.47^{\circ}\text{F}$

$T_{\text{model, initial}} = 89^{\circ}\text{F}$

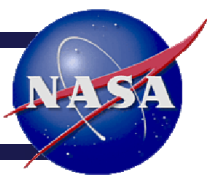
$\alpha = 2^{\circ}$



**Horseshoe Vortex at
Wing-Body Junction**



Turbulent regions cooler than laminar regions

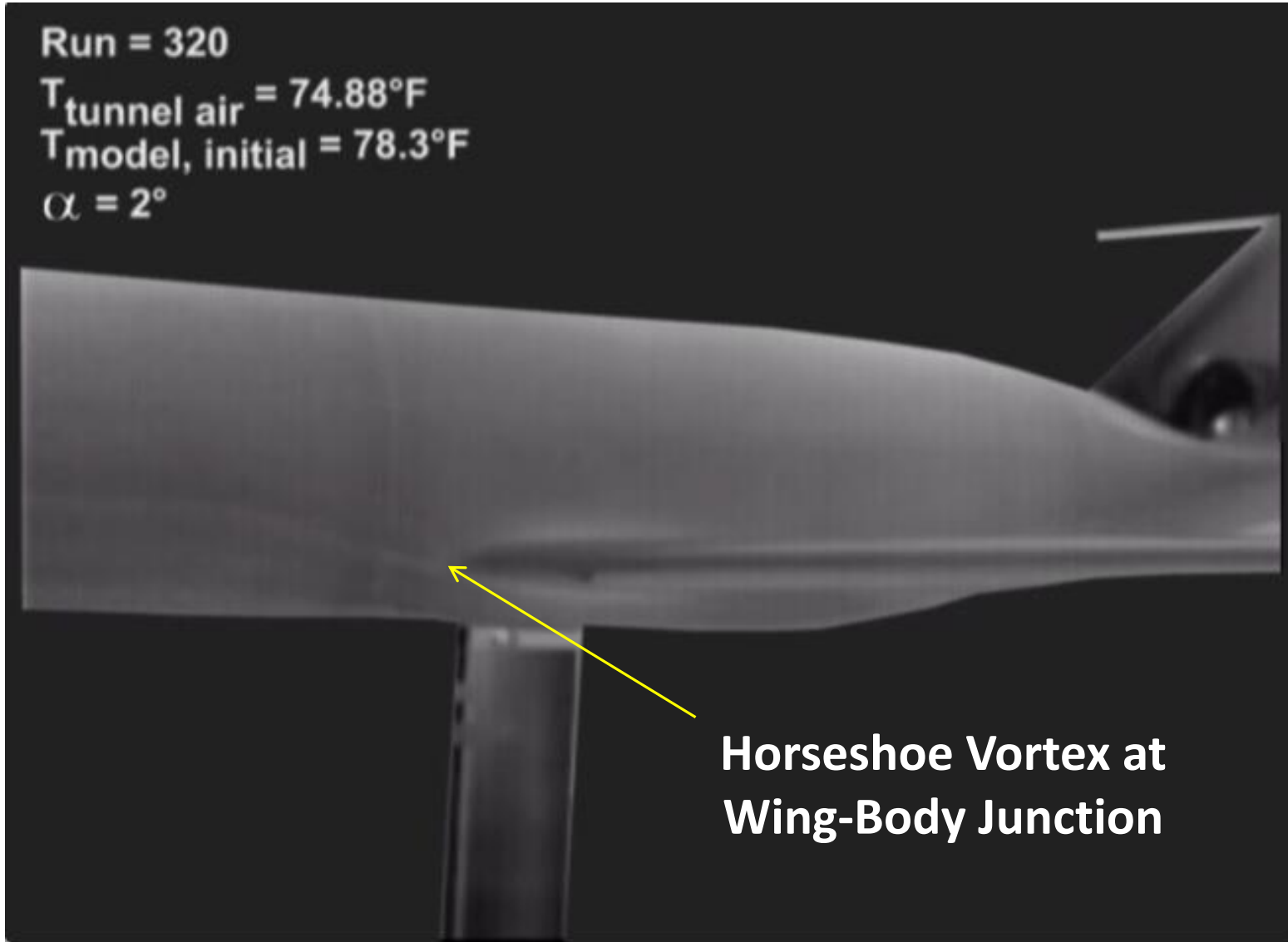


Run = 320

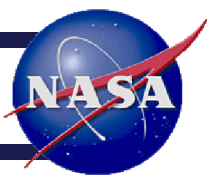
$T_{\text{tunnel air}} = 74.88^{\circ}\text{F}$

$T_{\text{model, initial}} = 78.3^{\circ}\text{F}$

$\alpha = 2^{\circ}$



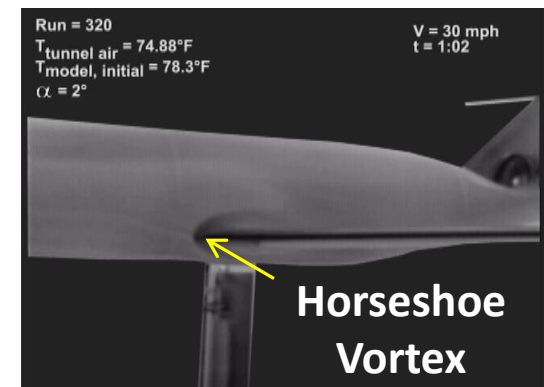
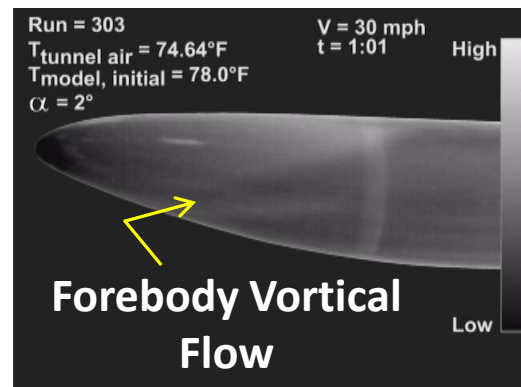
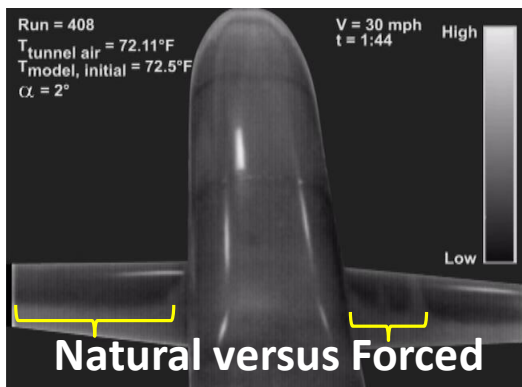
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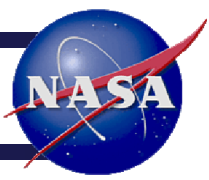


Conclusion – Lessons Learned

- A sufficient temperature difference between the wind tunnel air and the model surface showed the location of boundary layer transition
- Boundary layer transition was visible regardless of whether the wind tunnel air or the model surface was cooler
- Flow characteristics such as a wing root horseshoe vortex or the presence of forebody vortical flows were visualized with IR data
- Active temperature control of the model or the air would enhance the usefulness of IR images, but is not necessary to observe boundary layer transition

Questions?

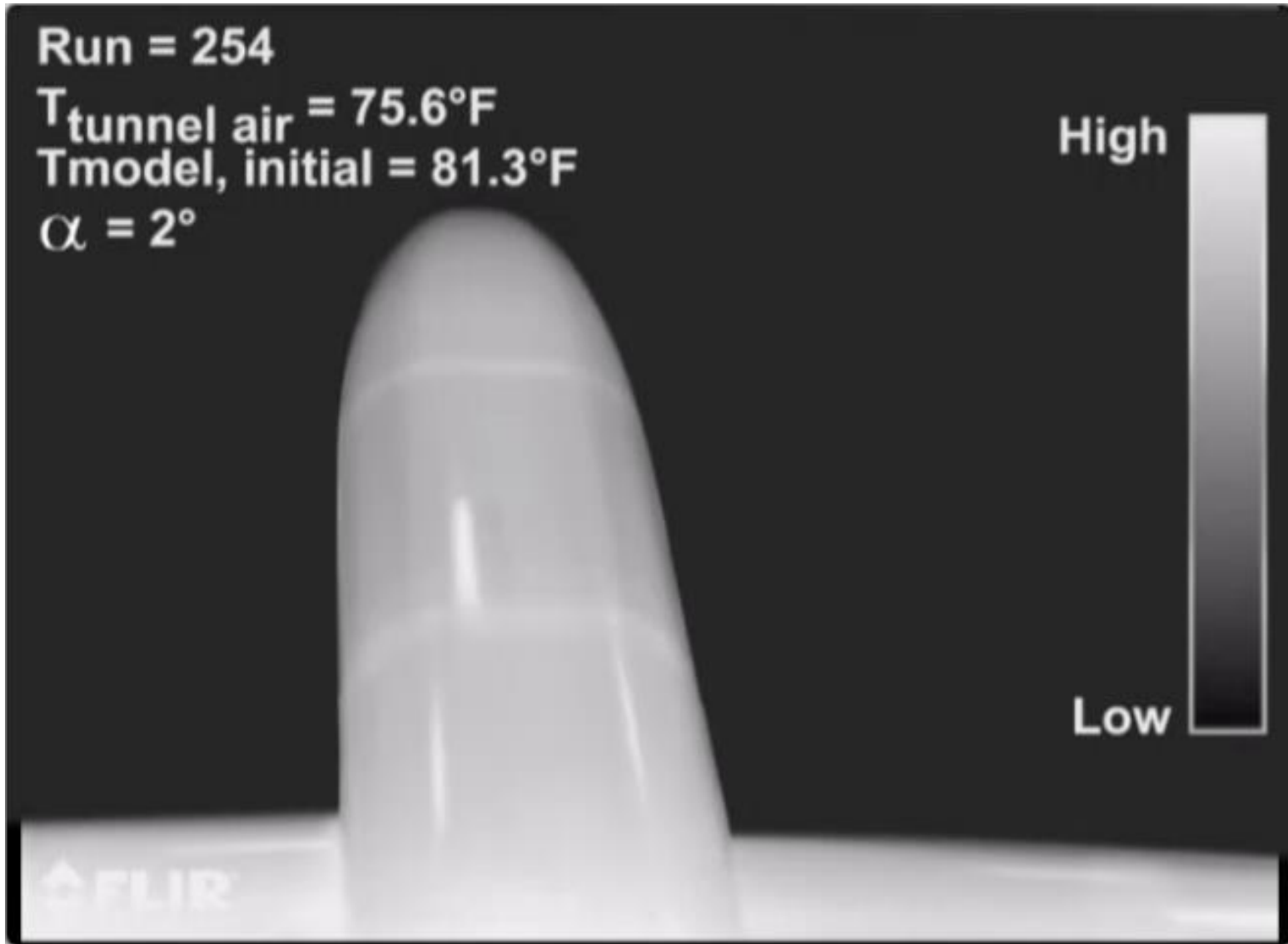
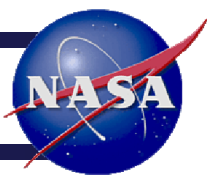




Back-Up Slides

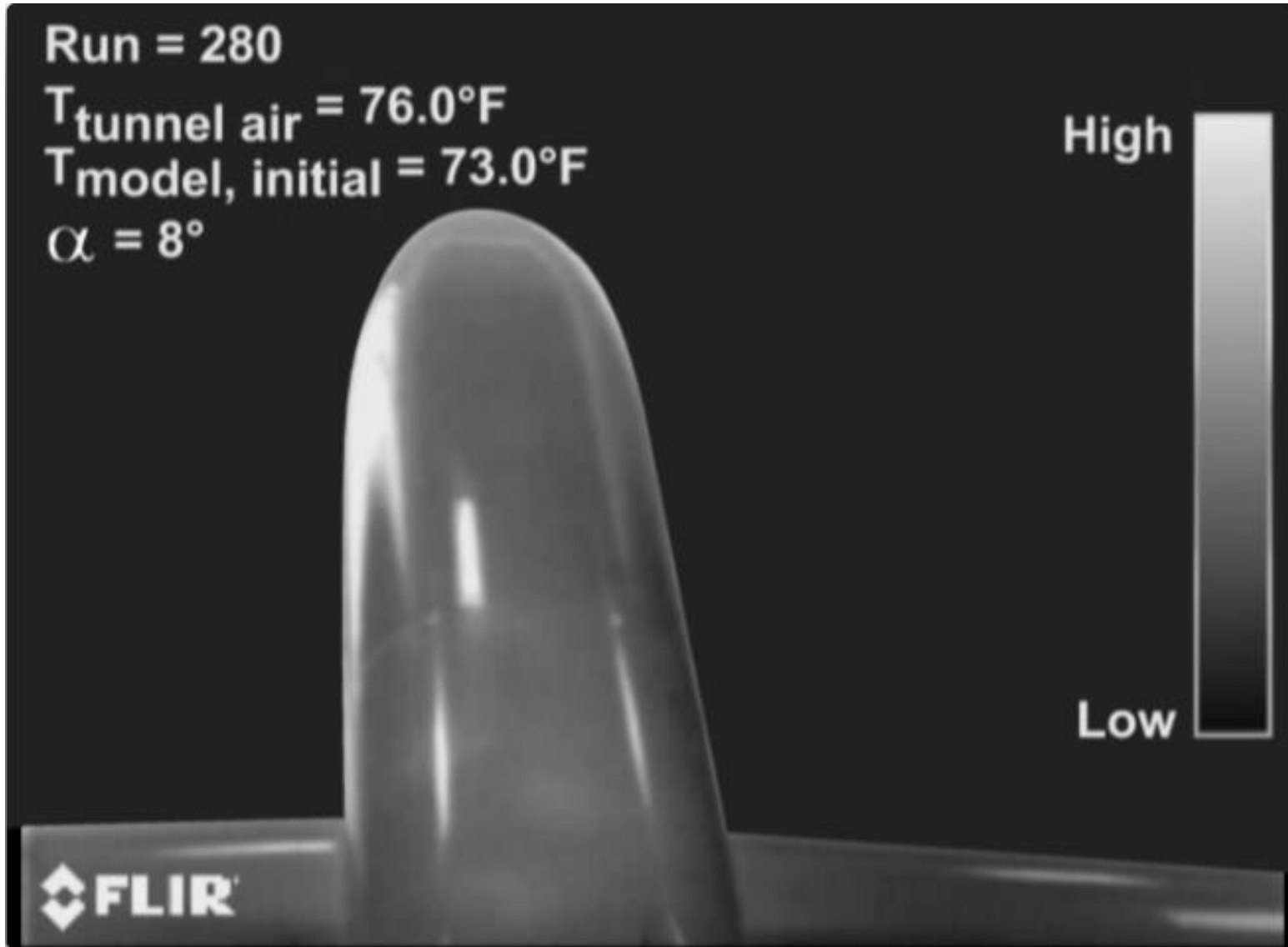
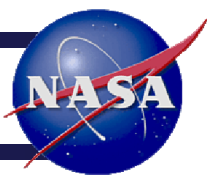


Turbulent regions cooler than laminar regions



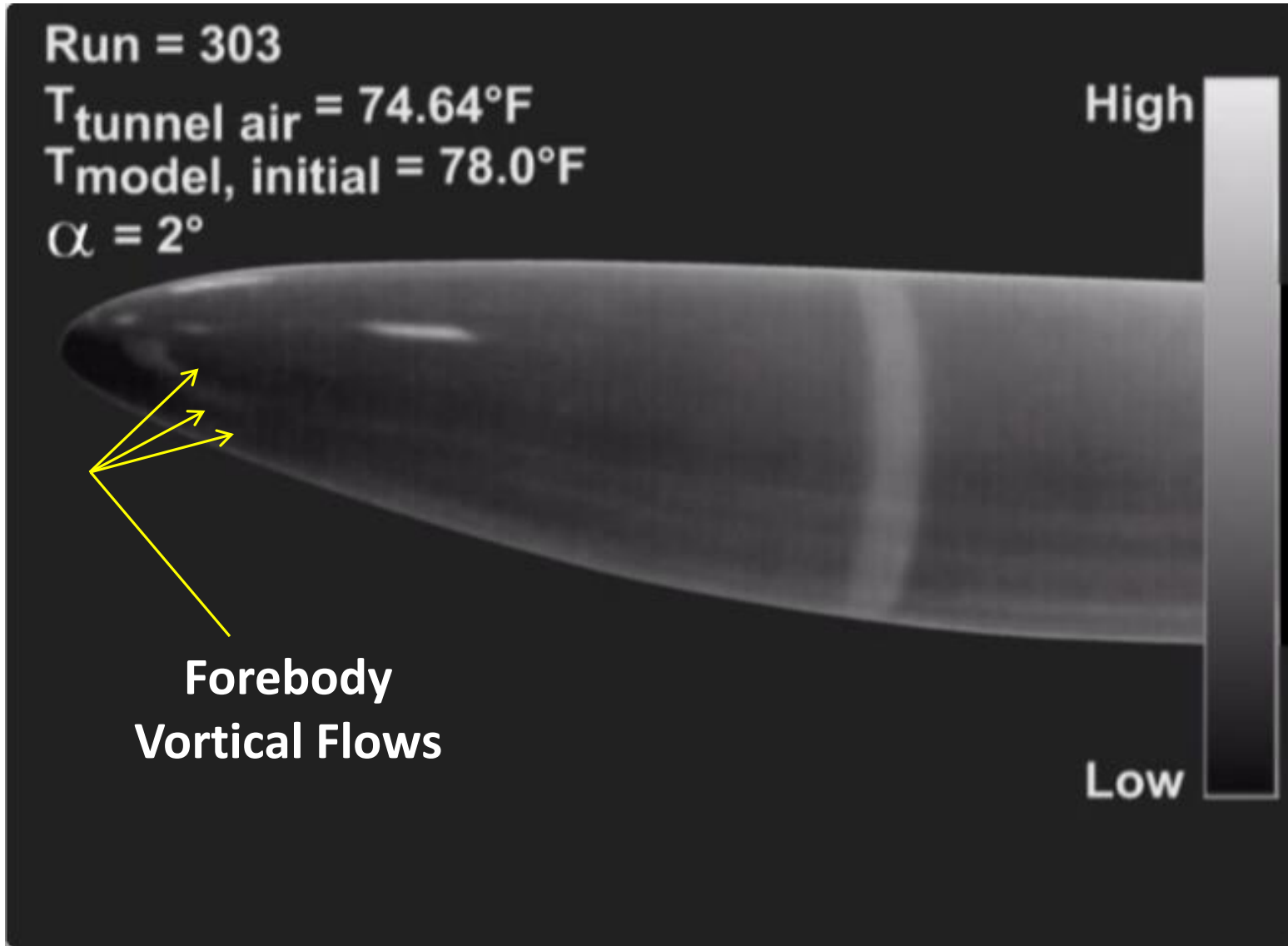
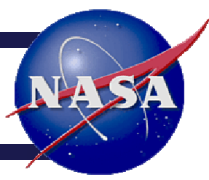


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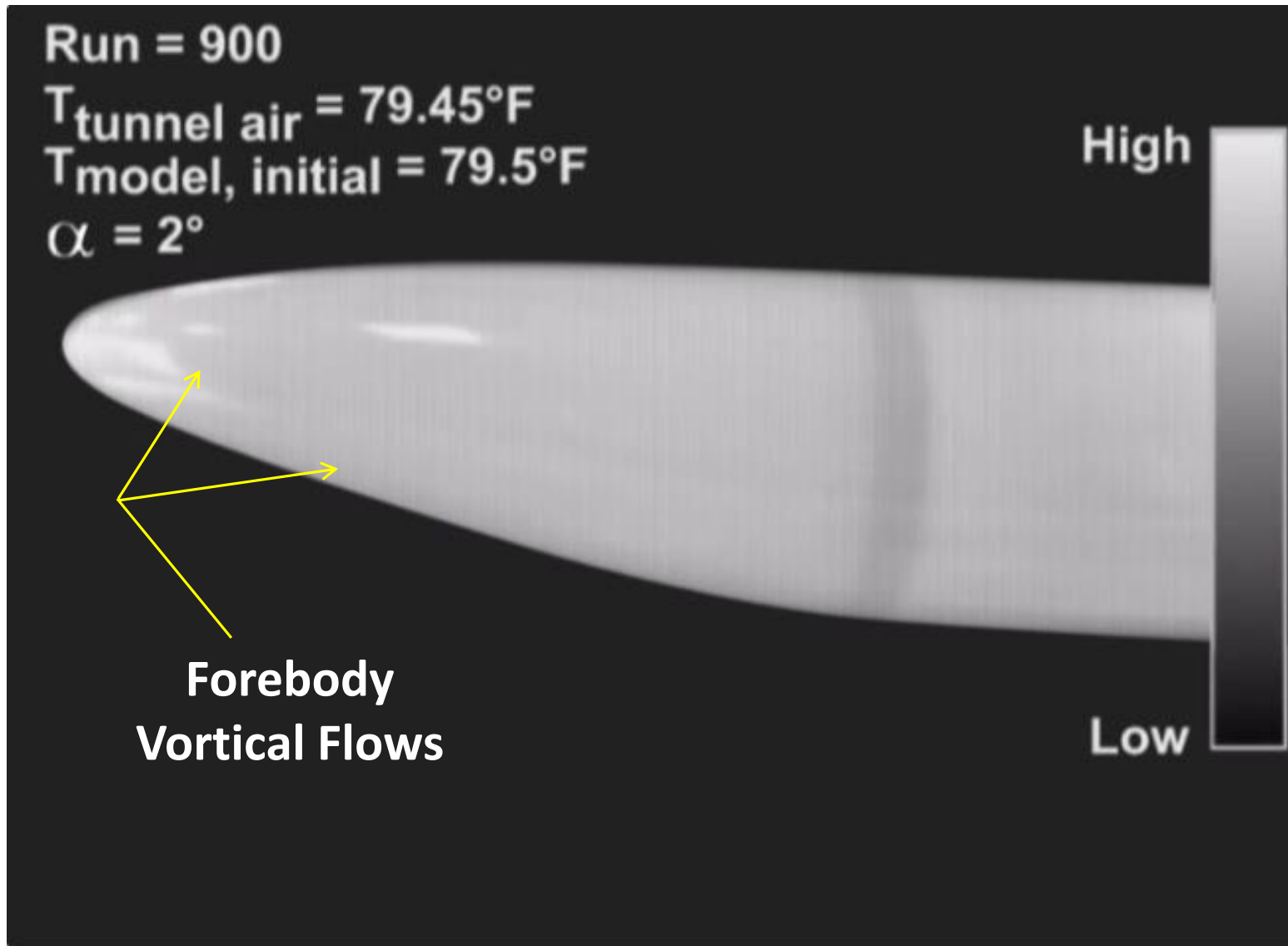


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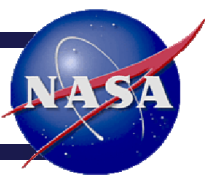


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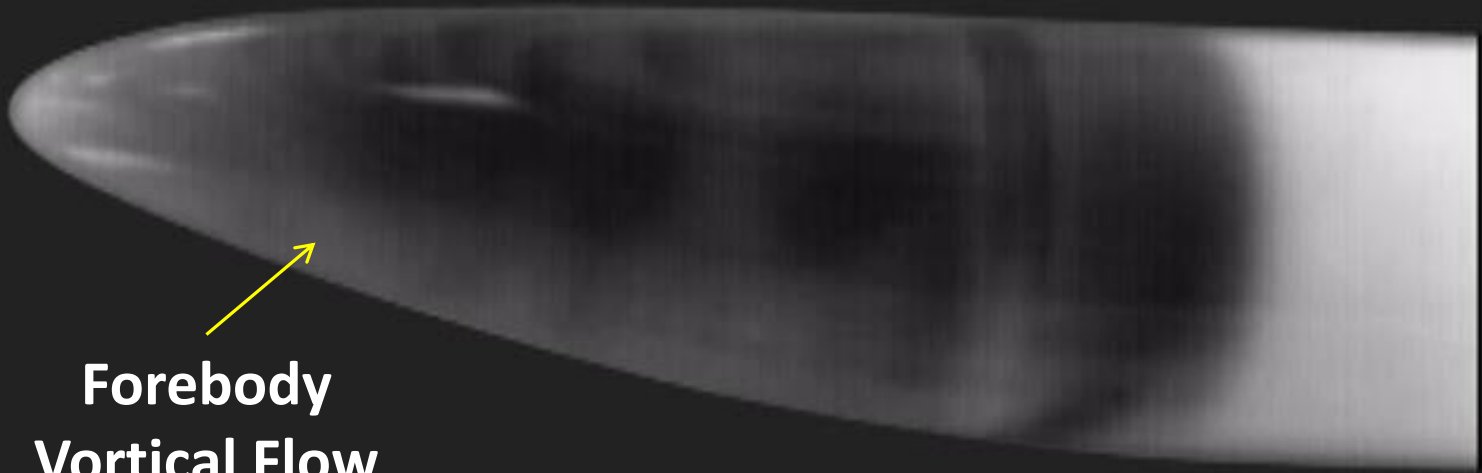


Run = 318

$T_{\text{tunnel air}} = 81.92^{\circ}\text{F}$

$T_{\text{model, initial}} = 79.3^{\circ}\text{F}$

$\alpha = 2^{\circ}$

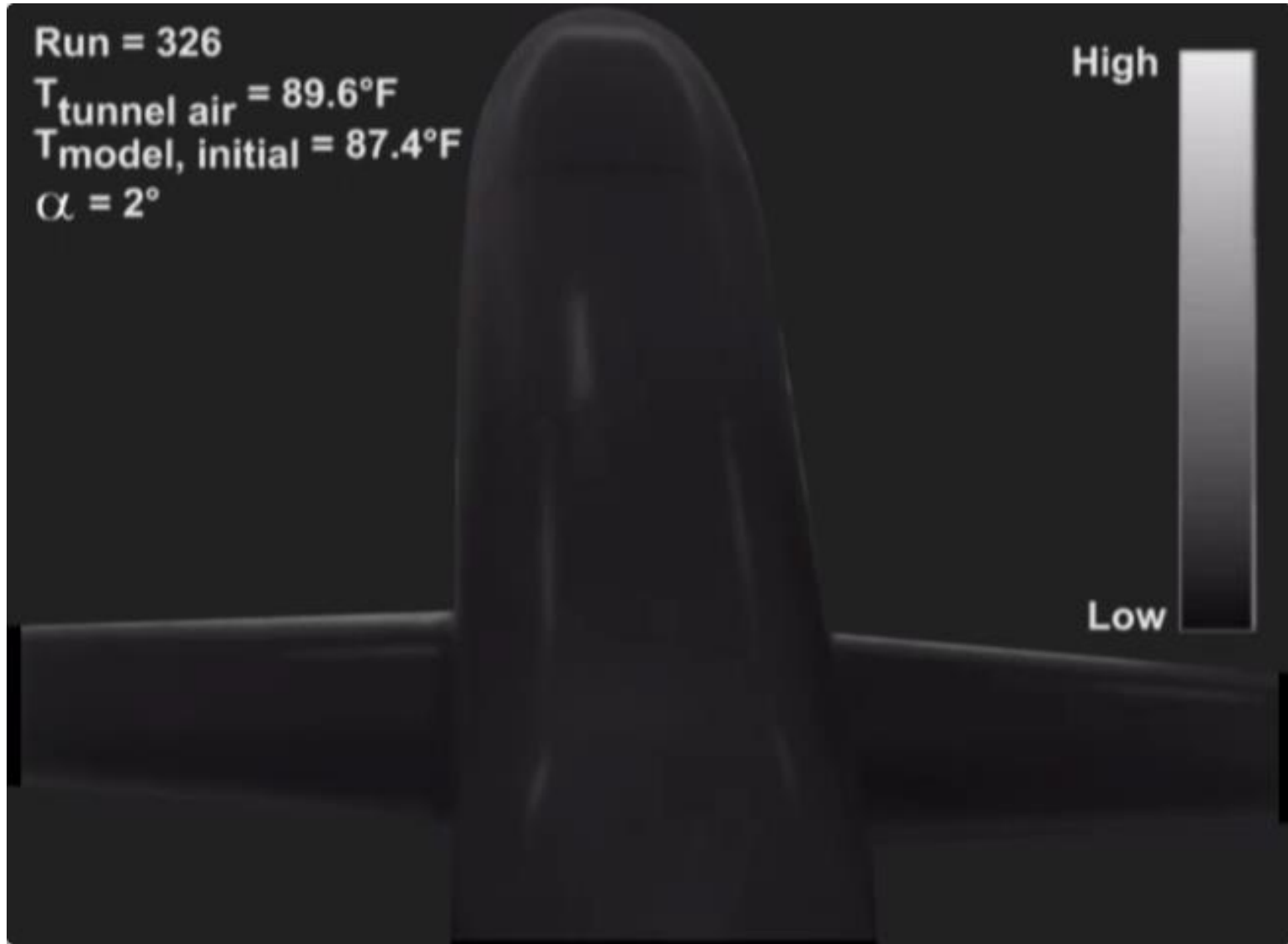
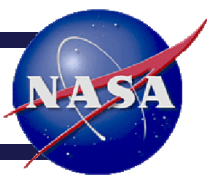


**Forebody
Vortical Flow**

**Model initially cooled with wet
towels on forward fuselage**

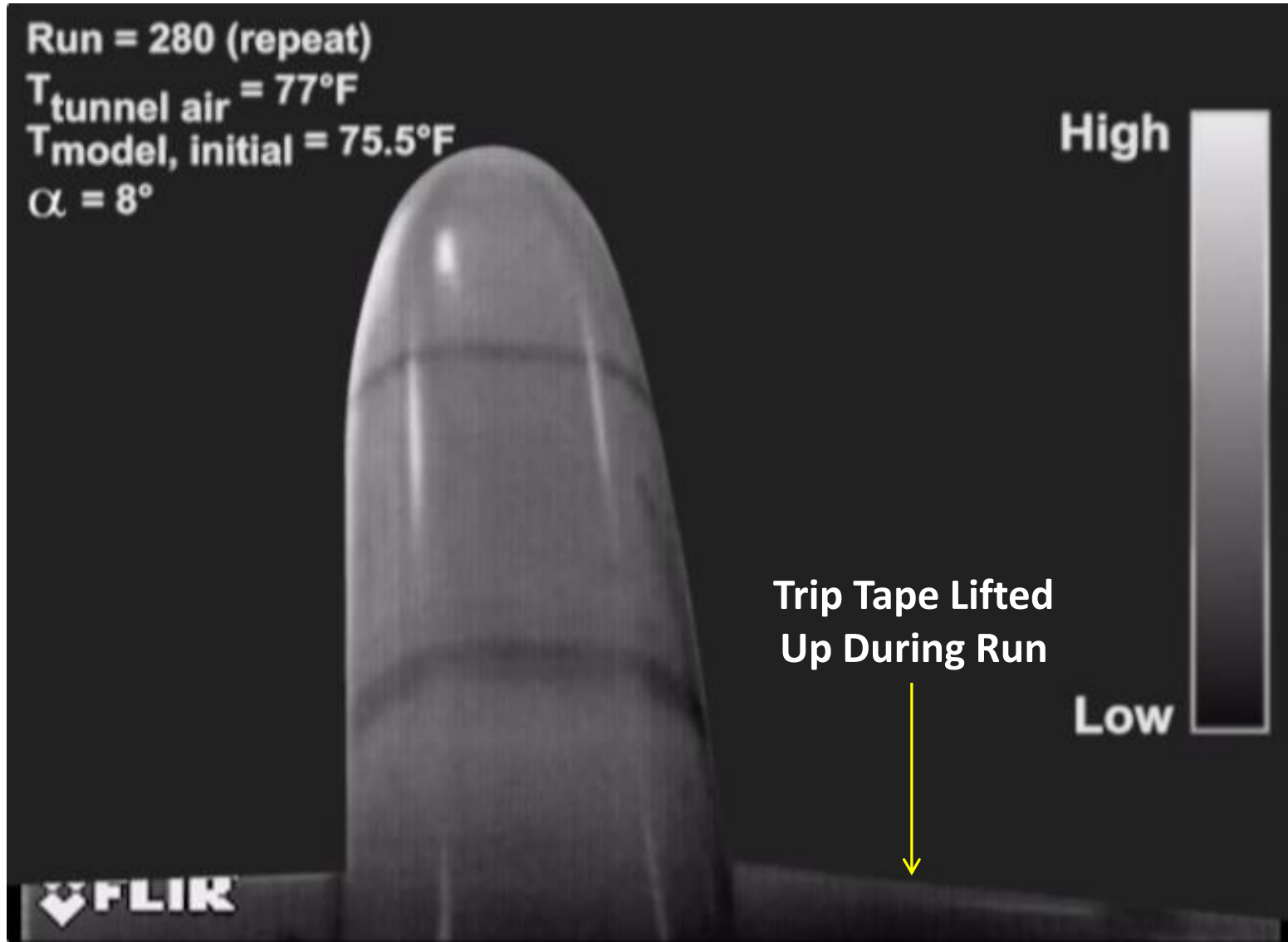
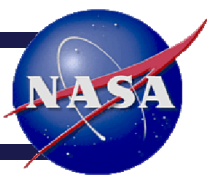


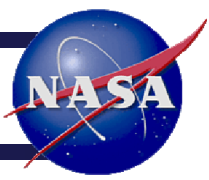
Natural and forced transition on wings



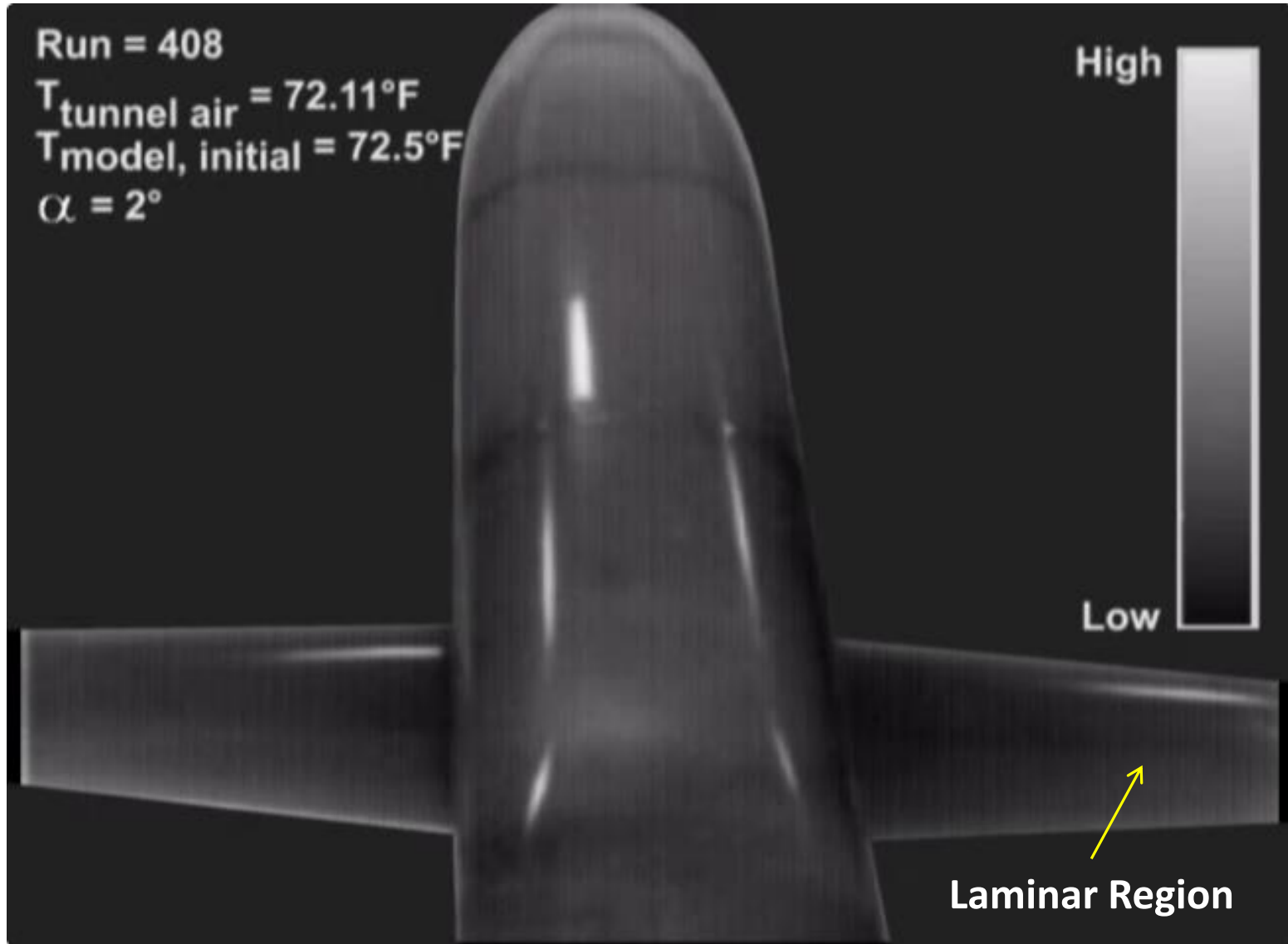


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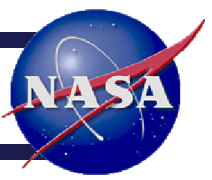


Laminar region behind removed trip tape





Turbulent regions warmer than laminar regions

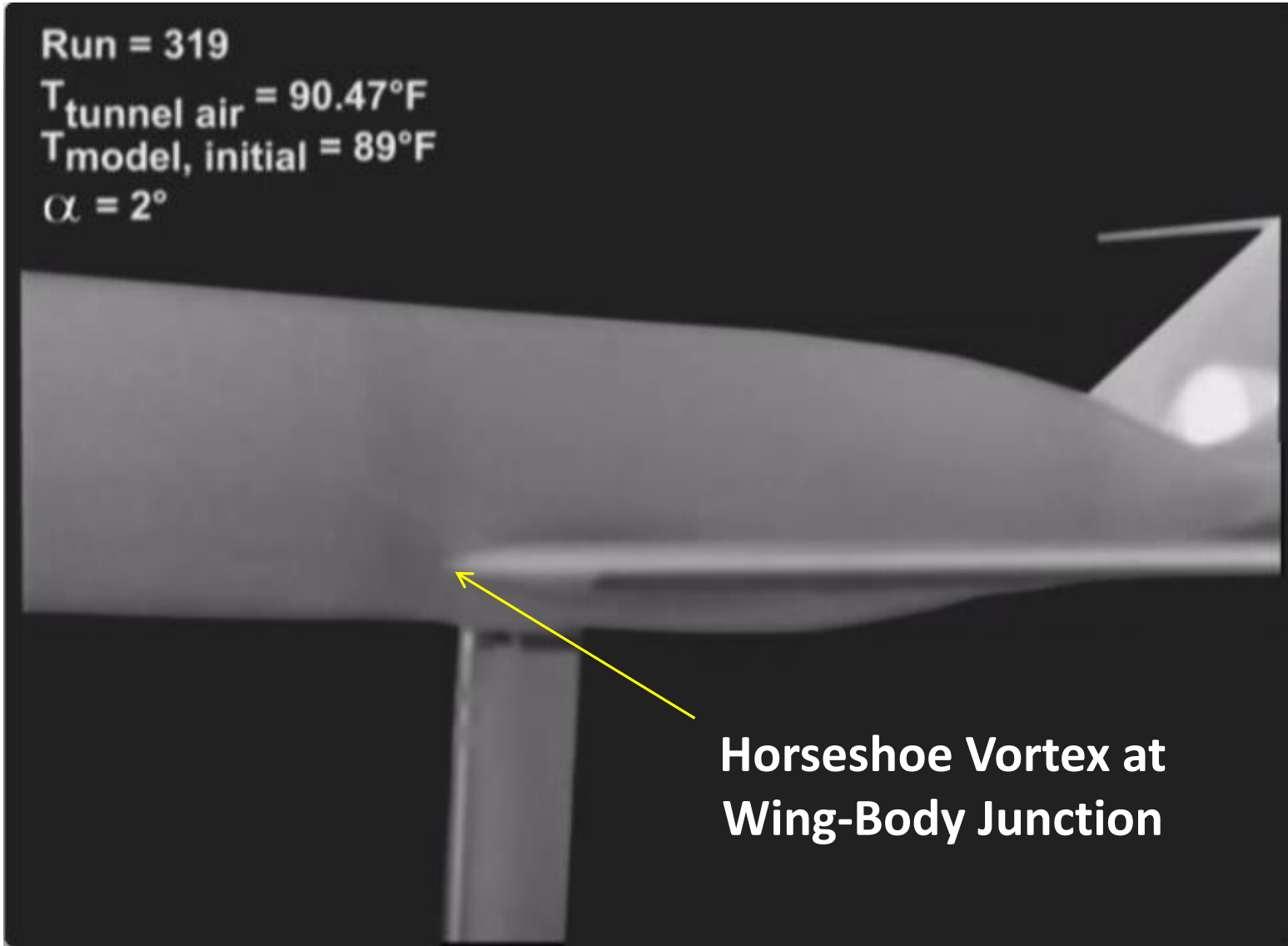


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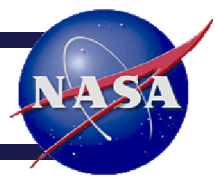
$\alpha = 2^{\circ}$



**Horseshoe Vortex at
Wing-Body Junction**



Turbulent regions cooler than laminar regions

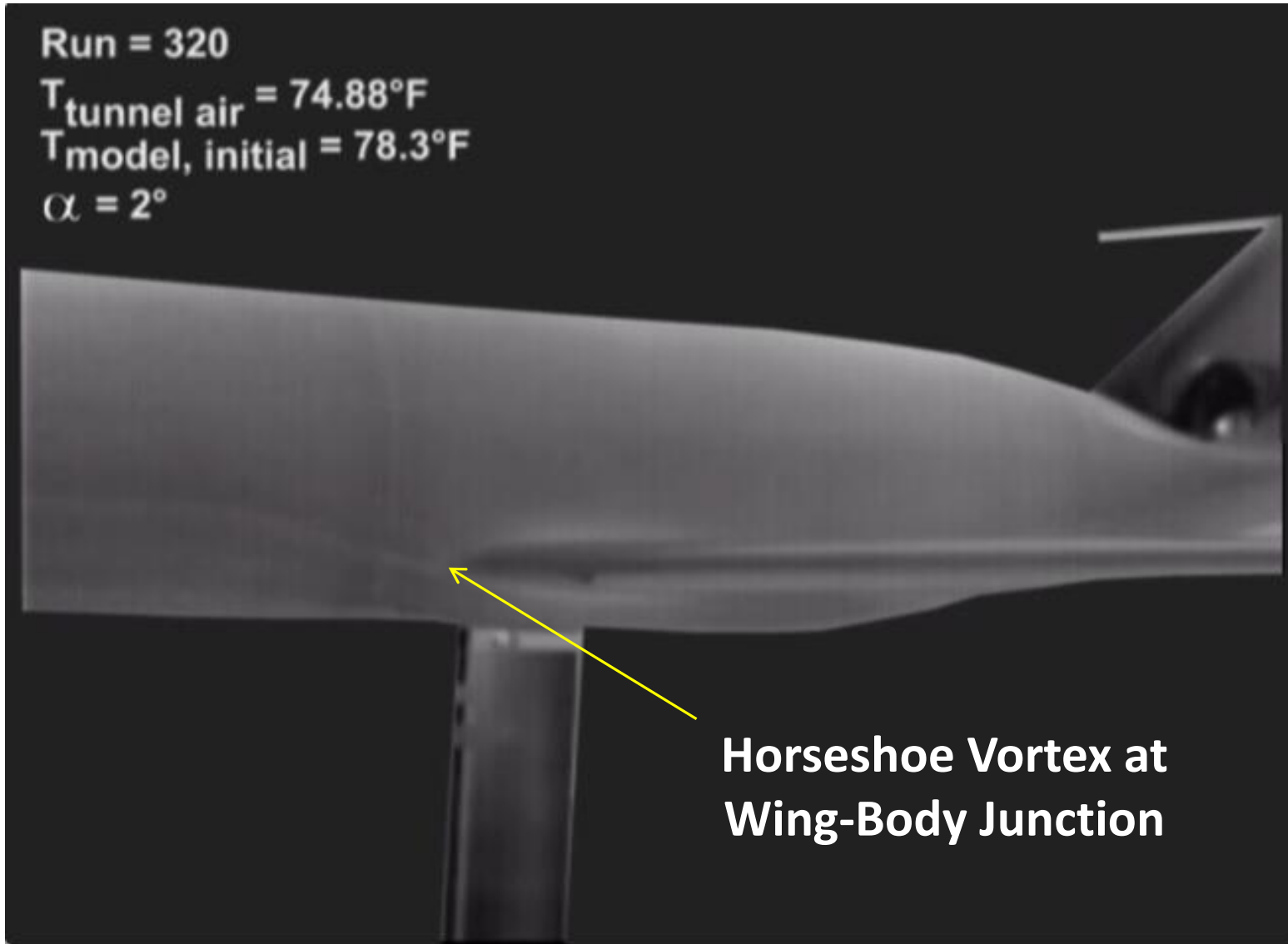


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