THERMAL BUCKLING OF CYLINDERS

By Melvin S. Anderson

NASA Langley *Research* **Center**

SUMMARY

Several theoretical **and experimental** investigations on the buckling **of cylinders** due **to both** axial and circumferential thermal **stresses** are **reviewed. Differences** that **exist** among **the various results** are **discussed** and areas of future **work** are indicated.

INTRODUCTION

In recent years numerous investigators have obtained theoretical solutions to various problems on the buckling of cylinders due to thermal stress and in some cases have obtained the interaction of thermal stresses with stresses due to externally applied loads. In the solution of these problems thermal stresses caused by arbitrary temperature distributions are generally represented by a Fourier series and then a Fourier series solution of the buckling problem is obtained. For a specific temperature distribution, the Fourier coefficients for the thermal stress are evaluated and the buckling conditions are determined. It can be seen that any varying stress distribution problem could be treated in the same manner; thus, the work that has been done could be transmitted to thermal stress problems. is not necessarily limited to thermal **stress problems.**

Two general cases **have been** considered **in** the **literature. The** The of a cylinder due to circumferential stresses that vary in the axial direction. Circumferential thermal stresses are caused mainly by restraint of thermal expansion in the vicinity of the cooler rings. The interaction of circumferential thermal stress with uniform axial compression is treated in reference 3. Buckling of a cylinder due to axial stresses that vary in the circumferential direction was consid**ared in references** 4 and 5. Such a stress distribution results when a cylinder is heated nonuniformly around the circumference. For combinations of these cases, in which both circumferential and axial stresses are varying, no theoretical solution is available, but cerstresses are vary included an theoretical solution of the separate analyses can tain conclusions based on the results of the separate analyses can be inferred.

In the throughout reference the throughout reference 12 to 5,000 km in the throughout reference 1.5, eventually references 1.5, $\frac{1}{2}$ Donnell equation **or** the modified Donnell equation (see ref. 6) was used.

Preceding page blank

 \boldsymbol{n} γl

Some of the more significant aspects of the various solutions are discussed in this paper.

 $\sim 10^{-11}$

 \sim ω

 $\overline{}$

 \hat{p} , \hat{p} , \hat{p}

SYMBOLS

 $\overline{}$

BUCKLING OF CYLINDERS DUE TO CIRCUMFERENTIAL STRESSES

THAT VARY **IN** THE AXIAL DIRECTION

When a ring-stiffened cylinder is heated rapidly, cooler rings cause
circumferential thermal stresses which are a maximum in compression over the rings and generally decay to zero away from the rings. The interaction of these thermal stresses with uniform axial compression to cause buckling is shown in figure 1. It is assumed that the rings and skin are both at constant temperatures, the difference in these temperatures being ΔT which is plotted as the ordinate in the form of the temperature buckling coefficient τ . The abscissa γ is the ratio of the ature buckling coefficient v. The abscribed buckling stress for a axial compressive stress to the classical buckling stress for **a** under uniform compression.

The solid curve from reference 3 applies to a cylinder **of** many bays reference 1 for the case of heating alone, except that the cylinder was reference \pm for the case of \pm from reference \pm considered to be one-bay long. The *value* of α is indicated by the circle symbol at γ equal to zero is about $2\frac{1}{2}$ times the

value obtained for a cylinder with many bays. This difference does not at first appear reasonable since the equations used in references l and 3 lead to the same stability determinant. However, as shown in figure 2 the thermal stress distribution is not the same for the two problems. In figure 2 the theoretical circumferential thermal stress $\frac{1}{\sigma_{xy}}$ is plotted in nondimensional form against $\frac{X}{L}$. The stress distri-

bution for a cylinder of many bays is given by the solid curve while the dashed curve applies to a one-bay simply supported cylinder. The extent of the region of compressive stresses is greater for the cylinder of many bays than for the one-bay cylinder. The reason for the difference between the two curves is that, in a cylinder of many bays, the radial expansion of the cylinder wall is symmetric about a ring which effectively clamps the ends of each bay as far as circumferential thermal **stress is concerned. For a given temperature, the average thermal stress** in a cylinder of many bays is over twice the average thermal stress for a cylinder of one bay. If the results of references 1 and 3 are compared on the basis of average stress at buckling, the cylinder of one bay is found to have an average stress only 20 percent greater than a cylinder found to have an average stress only 21 percent greater the buckling te of many bays, **even** though, as shown in figure l, the buckling tempera-

ture for a cylinder of one bay is over $2\frac{1}{2}$ times that for a cylinder of many bays. On the basis of this explanation, the result is reasonable and is similar to that found for buckling of flat plates where the averand is similar to that found for $\frac{1}{2}$ for $\frac{1}{2}$ at $\frac{1}{2}$ concentrates is concentrated when age stress at buckling increases somewhat mere toward the **edges.**

The curves in figure 1 indicate that a significant portion **of** the **classical** buckling **stress** can be applied without lowering the buckling temperature. **This** behavior can be **explained** by **examining** again the average circumferential **stress** in **the cylinder.** For a given tempera**ture,** it can be shown **(see** fig. 8 **of ref.** 3) **that** the **addition of** axial compression **reduces** the average circumferential thermal stress. For a **simply sup_rted** cylinder **(solid** curve in fig. 1) **the reduction** in average thermal **stress compensates** for the destabilizing **effect of** the axial compression and results in **essentiall_ no change of** the buckling temperature. For a clamped cylinder (long-dash-short-dash curve in fig. 1), this **compensation** is even greater so there is a **net** temperature increase for buckling. At high values of **7,** the axial stress becomes the **dominant** factor in buckling and the temperature rise at buckling **quickl_ drops** to zero at **7** equal to 0.95. This **5-percent** reduction in buckling stress from the classical value is caused by circumferential stresses produced by the restraint of Poisson's expansion in the vicinity **of** the rings_

An analysis **of** the buckling temperature **of** a clamped **end** cylinder is also given in reference **2. The** result **of** this analysis is indicated by the square **symbol** in figure **1** and is **seen** to be **almost** twice the value obtained **in** reference 3 **(long-dash--short-dash** curve **at 7** equal to **zero).** The stress distribution and the **boundary** conditions were the same in the two references; **however,** the **eighth-order Donnell** equation was used in reference 2 whereas the fourth-order modified Donnell equation was used in reference 3. A preliminary check of the results of reference **2** indicates there **ma_** be **some numerical errors** in the calculations; **however,** the correct **nmmerical** results obtained from the Donnell equation would still be **significantly** higher than those obtained in reference 3. **Batdorf** (ref. 6) indicated **that** the **use** of the Donnell equation in combination with a **Galerkin** solution, as was **done** in refer**ence 2,** could lead to incorrect results for clamped **cylinders.** The **dis**crepancy is possibly **due** to divergent series resulting from **differen**tiating the **deflection** function 8 times. Additional information on the accuracy of the Donnell equation for clamped cylinders is obtained from calculations made for a clamped cylinder loaded **to produce** a uniform circumferential stress, which is a limiting case of the more general thermal stress problem. **For** this case the result obtained from the Donnell equation is as **much** as **50** percent higher than either the result obtained by Sturm (ref. 7) or the result obtained from the modified **Donnell** equation. **The** reason for the **differences** between the two equations has **not been** completely **explained,** but it does appear that the **Donnell** equation gives incorrect results for clamped cylinders and that the modified equation should be **used.**

It is well known that **experimental** buckling results **are** lower than values calculated by **using small-deflection** theory for cylinders in axial

compression. However, for circumferential compression, theoretical
results are in agreement with tests. Thus, one could expect good agreement between theory and experiment at lower values of γ where circumferential compression is dominant but at higher values of γ the theory will be unconservative. The test points in figure 1 at τ equal to zero are for room-temperature bending tests, reported in reference 8, **zero are** for **roan-teml_erature** bending tests, **reported** in reference 8, **of** 7075-T6 aluminum-alloy cylinders **and** indicate **the** magnitude **of** the

reduction in buckling stress from the classical value for $\frac{r}{t}$ equal
to 300. In reference 3 it is proposed that γ be reduced at any value of τ by the same percentage as the reduction at room temperature. This procedure results in the dashed curve in figure 1. The two test points for the heating tests are from the results of reference 9 for 2024-T3 aluminum-alloy cylinders loaded in pure bending and then heated. $22 + T$ aluminum-alloy cylinders lotings the tests and the dashed curve. Reasonable agreement is shown between the tests and the **dashed** curve.

BUCKLING OF CYLINDERS DUE TO AXIAL STRESSES

ĥ,

THAT VARY IN THE CIRCUMFERENTIAL DIRECTION

If a cylinder is heated such that the temperature varies around the circumference, nonuniform axial thermal stresses will arise. The effect of these thermal stresses on the load-carrying ability of the cylinder can be inferred from the analyses given in references 4 and 5. It was shown in reference 4 that, based on small deflection theory for pure bending of a cylinder, the maximum compressive stress for practical cylinder proportions is essentially the classical buckling stress for a cylinder in uniform compression. For most cases of nonuniform heating, the axial thermal stress distribution in a cylinder is similar to a bending-moment stress distribution in the vicinity of maximum compressive stress. Thus, the theoretical maximum axial compressive thermal stress at buckling can also be taken as the classical buckling stress for uniform compression. An exception occurs if the region of compressive stress is very small in the circumferential direction. For example, in reference 10 a theoretical buckling stress slightly higher than the reference io a unconstant for a ovided heated along a very nar classical value is indicated for a cylinder **heated** along a very **narrow longitudinal** strip.

In reference 5, buckling **of** a cylinder under a **warying** axial stress Fourier cosine series and the resulting stability determinant is evaluated with the aid of certain simplifying assumptions. It is shown that the maximum stress for a bending distribution (cos φ) or the distribution given by the next higher term (cos 2p) approaches the classical value as the size of the determinant is increased. It is of interest to value as the size of the determinism $\frac{1}{2}$ increased. It is increased the stability note that with the use of the same simplifying $\frac{1}{2}$

259

 \checkmark

determinant for these two cases can be put in closed form, and the **maxstress** at buckling can be shown to be precisely the classical buckling stress for uniform compression when the **size** of the determinant approaches infinity.

It has **been mentioned** previously that theoretical buckling **stresses** for cylinders **must** be reduced to correspond to values observed in tests. **Thermal stress** distributions are **more** likely to be similar to bending stress **distributions** than uniform compression so that the **experimental** results obtained in bending tests of cylinders should **be** a good estimate of the maximum thermal **stress** at cylinder buckling. Experimental buck**llng stresses** for uniform compression, which are **slightly** lower than the results obtained in bending tests, would provide a conservative estimate of the thermal buckling stress.

Even though the maximum **stress** at cylinder buckling has been **speci**fied, it is still somewhat of a problem to **determine** the temperature **at** which this **stress** occurs. The usual **elementary** analysis **of** thermal stress will **not** be **sufficient** in many cases as is illustrated in figure 3 where the longitudinal and circumferential variation of axial ther**mal** stress is **shown** for a typical heating condition. Figure 3 was obtained from the results of reference 11 where an analysis is given of the axial thermal **stress** present in a cylinder with **a nonuniform** temperature distribution. **The dashed** curves represent the **elementary** ther**mal stress** in each bay, and they **differ** considerably from **either** the experimental points or the solid curves, which were calculated by the theory **of** reference ii. The thermal **stresses** shown in figure 3 were the result of heating a cylinder on **one** side only over the central portion **of** the cylinder. Strains were **not** measured in the areas directly under the **heating** lamps but, in the other regions of the cylinder, stresses **determined** from measured strains agree reasonably well with theoretical values.

As previously mentioned, the results **of** references 4 and 5 indicate that, if the classical buckling **stress** for uniform compression is acting over only **a** portion **of** the circumference **of** a cylinder, buckling still occurs. Hence, it is reasonable to assume that the interaction curve obtained in reference **5** for uniform heating and uniform compression would also apply to stresses acting over a portion of the cylinder circumference. Using this assumption and the results of the various buckling analyses that **have** been previously discussed, the experimental results of reference ll can be correlated with theory as shown in figure 4. In figure 4 , the maximum cylinder bending stress σ_X is plotted against maximum cylinder temperature Tmax. The **upper** curve is the result that would be obtained if there were no thermal stress and the bending strength was reduced in proportion to the reduction of Young's modulus with temperature. The middle curve was calculated from reference 3 for

a cylinder that is uniformly heated. The curve represents the inter-
action of a uniform compressive stress with the circumferential thermal stresses present due to the cooler rings. The cylinders represented by the test points on figure 4 were not heated uniformly but were heated as indicated in figure 3; this heating caused axial thermal stresses in addition to the load-induced stress. The maximum compressive bending stress was at the bottom of the cylinders which was also the point of maximum heating and high compressive thermal stresses; therefore, this region was critical for buckling. The load-carrying ability of the cylinder is indicated by the lower curve which represents the allowable load-induced stress and was obtained by subtracting the calculated axial thermal stress at the bottom of the cylinder from the value given by the middle curve. The data exhibit scatter which is about the same as middle curve. The data exhibit scatter which is about the same as observed in the results of room-temperature tests and is in reasonable agreement with predicted values.

AREAS OF FUTURE RESEARCH

For the problem **of** buckling due to circumferential thermal stress, emounts of axial compression have also been present. Further research is needed in this area to determine experimental buckling results for cylinders that buckle due to temperature alone. In addition the loadcylinders that buckle due to temperature and index the ld be of inte carrying ability of the temperature-buckled control

Axial thermal stresses could be a significant factor in buckling **of** structure of launch vehicles. Therefore, it would appear that extension of the method of reference 11 for calculating thermal stresses to include stiffened cylinders with the possibility of local buckling between stiffeners should be an area for future research. Also the effect of a varying axial stress on the buckling behavior of a longitudinally stiffened cylinder should be investigated in order to determine whether there is any der should be investigated in the seal of uniform compression. significant difference from the case of uniform compression.

CONCLUDING REMARKS

The work of several investigators on the buckling of cylinders under varying axial and circumferential thermal stresses has been reviewed. It has been shown that the severity of the circumferential thermal stress is strongly dependent on the boundary conditions. The interaction of circumferential thermal stresses with axial compression is also discussed. cumierential thermal stresses with a complete $\frac{1}{2}$ in consequential conduction is A method of incorporating these results with the observed reduction in the contract of the observed reduction in the c

axial buckling stress from the theoretical value is indicated and is shown to agree with experimental results.

<u>PARAS ARAS LO DA OFFICIALIST</u>

For cylinders that are heated nonuniformly, the axial thermal stress cannot in general be predicted by elementary theory. A method is available to predict these stresses which can be added to the load-induced For cylinders that **are heated nonuniformly,** the axial thermal **stress** cannot in general be **predicted by elementary theory. A method** is **avail**to the buckling stress that would be obtained in a cylinder heated uniformly and loaded by bending or compression.

REFERENCES

- 1. Hoff, **N.** J. **: Buckling of Thin** Cylindrical **Shell Under** Hoop Stresses **Varying in** Axial **Direction. Jour. Appl. Mech., vol. 24, no.** 3, **Sept.** 1957, **PP.** 405-412.
	- **2. Johns, D. J.,** Houghton, D. S., **and Webber, J. P.** H.: **Buckling Due TO** _hermal Stress of **Cylindrical Shells Subjected** to **Axial Temperature Distributions. Rep.** No. **147, The College** of **Aeronautics, Cranfield** (British), **May** 1961.
- 3. Anderson, **Melvin S.** : Combinations of Temperature and **Axial** Compres**sion Required** for **Buckling** of a **Ring-Stiffened Cylinder.** NASA **TN D-1224, 1962-**
	- 4. Seide, Paul, and Weingarten, V. I.: On the Buckling of Circular Cylindrical Shells Under Pure Bending. Trans. ASME, Ser. E Jour. Cylindrical Shells **Under Pure** Bending. Trans. _, **Ser.** E **-** Jour. Appl. **Mech.,** vol. 28, **no.** l, **Mar. 1961** s PP. **112-116.**
	- **5.** Abir, David, Hoff, N. J., Nardo, **S. V.,** Pohle, **Frederick** V., Vafakos, William, **and** Wan, **Eoon-Sang:** Thermal Buckling of Circular **Cylindrical and Conical Thln-Walled** Shells. **WADC Tech.** Rep. **58-I04** ASTIA **Doe. No.** AD **151068, U.S.** Air Force, Apr. **1958.**
- 6. Batdorf, S. B. : A **Simplified** Method **of** Elastic-Stability Analysis for $\frac{1}{2}$ $\frac{1}{2$ included in NACA $\overline{1}$ s $13#1$ and $13#2.$)
- 7. Sturm, Rolland G. : A **Study** of the **COllapsing Pressure of** Thin-Walled Cylinders. Eng. Exp. Stat. Bull. No. 329, **1941.**
	- **8.** Peterson, **James** P. : Bend/rig **Tests** of **Ring-Stiffened** Circular Cyllnders. **NACA** TN 3735, **1956.**
- 9. **Pride,** Richard A., Hall, John **B.,** Jr., and Anderson, Melvin **S.:** Effects of **Rapid** Heating on Strength **of** Airframe Components. NACA TN 4051, 1957.
- i0. Rill, D. **W.** : Buckling of a Thin **Circular** Cylindrical Shell Heated Along an Axial **Strip. SUDAER No.** 88 (AFOSR-TN-99-1250), Dept. Aero. Eng., Stanford Univ., Dec. **1959.**
- L. Anderson, Melvin S., and Card, Memerican a Nonuniform Temperati Cylinders Under **a Pure** Bending Moment and **a** Nonuniform Temperature Distribution. NASA TN D-1513, 1962.

 \sim . \sim

Figure 1

VARIATION OF CIRCUMFERENTIAL THERMAL STRESS
ALONG BAY LENGTH

Figure 2

 $\mathbf{\hat{z}}$

Figure 3

Figure 4

 \blacksquare

 $\frac{1}{2}$, $\frac{1}{2}$ $2^{\lfloor \sqrt{k} \rfloor}$

 $\label{eq:2.1} \frac{1}{\sqrt{2}}\int_{\mathbb{R}^3}\frac{1}{\sqrt{2}}\left(\frac{1}{\sqrt{2}}\right)^2\frac{1}{\sqrt{2}}\left(\frac{1}{\sqrt{2}}\right)^2\frac{1}{\sqrt{2}}\left(\frac{1}{\sqrt{2}}\right)^2\frac{1}{\sqrt{2}}\left(\frac{1}{\sqrt{2}}\right)^2.$ \mathcal{L}^{max}

 $\mathcal{L}_{\mathrm{max}}$

 $\label{eq:2.1} \frac{1}{\sqrt{2}}\int_{\mathbb{R}^3}\frac{1}{\sqrt{2}}\left(\frac{1}{\sqrt{2}}\right)^2\frac{1}{\sqrt{2}}\left(\frac{1}{\sqrt{2}}\right)^2\frac{1}{\sqrt{2}}\left(\frac{1}{\sqrt{2}}\right)^2\frac{1}{\sqrt{2}}\left(\frac{1}{\sqrt{2}}\right)^2.$