ANALYTICAL INVESTIGATION OF
SUPersonic TURBOMACHINERY BLADING

I - Computer Program for Blading Design

by Louis J. Goldman and Vincent J. Scullin

Lewis Research Center
Cleveland, Ohio
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NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

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ANALYTICAL INVESTIGATION OF SUPERSONIC TURBOMACHINERY BLADING

I - COMPUTER PROGRAM FOR BLADING DESIGN

by Louis J. Goldman and Vincent J. Scullin

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SUMMARY

A FORTRAN IV computer program for the design of supersonic blading based on establishing vortex flow within the blade passage is presented. The method of characteristics, as applied to the two-dimensional isentropic flow of a perfect gas, was utilized for the blade design. The equations necessary for the design are developed. The information required for the program consists of an inlet flow angle, specification of the inlet, outlet, and lower- and upper-circular-surface Mach numbers, and the specific-heat ratio. The program output consists of the blade coordinates and, if desired, a printer plot of the blade profile and flow passage. In addition, supersonic starting and flow separation calculations are performed by the program and obtained as output. An example is included to indicate the use of the program and the results obtainable.

INTRODUCTION

Supersonic compressors and turbines are employed in special circumstances because of their simplicity and low weight. A recent application for a supersonic turbine involves the hydrogen-fueled open-cycle auxiliary space power system described in reference 1. If the highest practical efficiency is to be obtained from supersonic compressors or turbines, proper design methods must be available.

A method for designing supersonic blade sections based on two-dimensional isentropic flow is given in reference 2. The method consists of converting the uniform parallel flow at the blade inlet into a vortex flow field, turning the vortex flow, and reconverting to a uniform parallel flow at the blade exit. The application of this design procedure involves specification of the inlet and outlet Mach numbers, the lower- (or concave) surface Mach number, the upper- (or convex) surface Mach number, the inlet flow
angle, and the specific-heat ratio of the working fluid. In general, a wide range of designs is possible by selection of these parameters. Guidance in the selection of a blade design is obtained by considering blade shape, solidity, and supersonic starting and flow separation problems. In reference 2 the effect of some of the design parameters for low Mach numbers and a specific-heat ratio of 1.4 is examined.

In view of the interest in hydrogen-fueled auxiliary space power systems, an analysis was conducted to gain a better understanding of the effects of the design parameters on the resulting blade geometry and to extend the results of reference 2 to levels of interest for such systems. In reference 3, the effect of surface Mach numbers, inlet flow angle, and specific-heat ratio on the geometric characteristics of supersonic impulse turbine-blade sections is investigated over an inlet Mach number range of 1.5 to 5.0.

Blade design limitations resulting from supersonic starting and flow separation problems are also considered. In the present report, a description and a FORTRAN IV listing of a computer program for the design of blading applicable for any supersonic Mach number level and specific-heat ratio are presented. Supersonic starting and flow separation calculations are also performed by the program. An example is included to indicate the use of the program and the results obtainable. The report is organized so that those persons desiring to use the program need only read the sections METHOD OF ANALYSIS, DESCRIPTION OF INPUT, and DESCRIPTION OF OUTPUT. All necessary information pertaining to the program itself is contained in the sections DESCRIPTION OF INPUT, DESCRIPTION OF OUTPUT, and PROGRAM DESCRIPTION.

SYMBOLS

A area, ft² (m²)
a speed of sound, ft/sec (m/sec)
C reduction in maximum weight flow due to two-dimensional flow (eq. (34b))
C* dimensionless blade chord, chord/r *
f(R*) function defined by eq. (10b)
G* dimensionless blade spacing, spacing/r *
g_c dimensional conversion constant, 32.17 ft-lb/(lb)(sec²) (1 kg-m/(N)(sec²))
h blade height, ft (m)
j index for upper surface of blade
K* dimensionless vortex constant defined by eq. (23)
\( K_{\text{max}} \) \hspace{1em} \text{value of } K^* \text{ for which weight flow is maximum}

\( k \) \hspace{1em} \text{index for lower surface of blade}

\( M \) \hspace{1em} \text{Mach number, } V/a

\( M^* \) \hspace{1em} \text{dimensionless velocity or critical velocity ratio, } V/V_{\text{cr}}

\( (M_f^*)_{\text{max}} \) \hspace{1em} \text{maximum inlet velocity ratio for supersonic starting}

\( (M_f^*)_{\text{min}} \) \hspace{1em} \text{minimum lower-surface velocity ratio from separation criterion}

\( (M_u^*)_{\text{max}} \) \hspace{1em} \text{maximum upper-surface velocity ratio from separation criterion}

\( m \) \hspace{1em} \text{slope of Mach line}

\( \overline{m} \) \hspace{1em} \text{slope of wall segment}

\( p \) \hspace{1em} \text{static pressure, } \text{lb/ft}^2 (\text{N/m}^2)

\( (p_l^*)_{\text{max}} \) \hspace{1em} \text{maximum lower-surface static pressure from separation criterion}

\( Q \) \hspace{1em} \text{vortex flow parameter (eq. 34a))}

\( R \) \hspace{1em} \text{radius in vortex field, } \text{ft (m)}

\( R^* \) \hspace{1em} \text{dimensionless radius in vortex field, } R/r^*

\( r^* \) \hspace{1em} \text{radius of sonic velocity streamline in vortex field, } \text{ft (m)}

\( V \) \hspace{1em} \text{velocity, } \text{ft/sec (m/sec)}

\( V_{\text{cr}} \) \hspace{1em} \text{critical velocity, } \text{ft/sec (m/sec)}

\( w \) \hspace{1em} \text{weight flow, } \text{lb/sec (kg/sec)}

\( w_{\text{max}} \) \hspace{1em} \text{maximum weight flow, } \text{lb/sec (kg/sec)}

\( X^* \) \hspace{1em} \text{dimensionless } X\text{-coordinate of blade (fig. 1), } X/r^*

\( x^* \) \hspace{1em} \text{dimensionless } x\text{-coordinate of transition arc (fig. 1), } x/r^*

\( Y^* \) \hspace{1em} \text{dimensionless } Y\text{-coordinate of blade (fig. 1), } Y/r^*

\( y^* \) \hspace{1em} \text{dimensionless } y\text{-coordinate of transition arc (fig. 1), } y/r^*

\( \alpha \) \hspace{1em} \text{circular arc turning angle, rad}

\( \beta \) \hspace{1em} \text{flow angle outside blade passage, rad}

\( \gamma \) \hspace{1em} \text{specific-heat ratio}

\( \theta \) \hspace{1em} \text{total flow turning angle, rad}

\( \mu \) \hspace{1em} \text{Mach angle, rad}
\( \nu \)  
Prandtl-Meyer angle, angle through which flow must turn from Mach 1 to required Mach number, rad

\( \Delta \nu \)  
incremental flow turning, rad

\( (\nu_1)_{\text{max}} \)  
magnitude inlet Prandtl-Meyer angle for supersonic starting, rad

\( (\nu_i)_{\text{min}} \)  
minimum lower-surface Prandtl-Meyer angle from separation criterion, rad

\( (\nu_u)_{\text{max}} \)  
maximum upper-surface Prandtl-Meyer angle from separation criterion, rad

\( \rho \)  
density, lb/ft\(^3\) (kg/m\(^3\))

\( \sigma \)  
blade solidity, \( C^*/G^* \)

\( \phi \)  
velocity direction angle, rad

Subscripts:

d  
downstream of normal shock

i  
blade inlet

j  
index for upper surface of blade

k  
index for lower surface of blade

l  
lower surface of blade

max  
maximum

min  
minimum

o  
blade outlet

u  
upper surface of blade

Superscripts:

total-state conditions

**METHOD OF ANALYSIS**

**Blade Description**

The design of supersonic blade sections described herein is based on establishing vortex flow within the blade passage by a procedure analogous to that given in reference 2. The blade so designed consists essentially of three major parts: (1) inlet transition arcs, (2) circular arcs, and (3) outlet transition arcs. A typical blade, with pertinent nomenclature noted, is shown schematically in figure 1. The inlet transition arcs
Figure 1. - Typical supersonic blade section. (All coordinates are made dimensionless by dividing by $r^*$.)

(lower and upper) are required to convert the assumed uniform parallel flow at the blade inlet into vortex flow. The concentric circular arcs turn and maintain the vortex flow. Finally, the outlet transition arcs reconvert the vortex flow into uniform parallel flow at the blade exit. Straight-line segments parallel to the inlet and outlet flow directions complete the blade profile. Methods for creating a finite thickness at the leading and trailing edges are given in reference 2.

In general, the inlet lower transition arc reduces the Mach number from its value at the blade inlet $M_i$ to a preselected value of the lower-surface Mach number $M_{l1}$, whereas the inlet upper transition arc increases the Mach number to a preselected value
of the upper-surface Mach number $M_u$. The surface Mach numbers remain constant, at these preselected values, on the lower and upper circular arcs. At the outlet region the procedure is reversed. The surface Mach number variation is shown in figure 2 for a typical blade.

The amount of flow turning produced by either the lower or upper surface of the blade consists, in general, of two parts (fig. 1): (1) the turning produced by the transition arcs and (2) the turning produced by the circular arcs. When isentropic flow turning at supersonic speeds is considered, it is convenient to introduce the Prandtl-Meyer angle $\nu$, which is defined as the angle through which the flow must turn from Mach 1 to the required Mach number. The flow turning produced by a transition arc is then equal to differences in Prandtl-Meyer angles and is $\nu_1 - \nu_\ell$ and $\nu_\circ - \nu_\ell$ for the inlet and outlet lower transition arcs and $\nu_u - \nu_1$ and $\nu_u - \nu_\circ$ for the inlet and outlet upper transition arcs, respectively. The turning produced by the inlet or outlet transition arcs cannot exceed the inlet or outlet flow angle $\beta_1$ or $\beta_\circ$, respectively. The relation between Prandtl-Meyer angle $\nu$ and Mach number $M$ is given by the following equations (ref. 4):
\[ \nu = \frac{\pi}{4} \left( \sqrt[\gamma-1]{\gamma+1} - 1 \right) + \frac{1}{2} \left\{ \sqrt[\gamma-1]{\gamma+1} \arcsin \left[ (\gamma - 1)M^* - \gamma \right] + \arcsin \left( \frac{\gamma+1}{M^*} - \gamma \right) \right\} \]  

where the dimensionless velocity or critical velocity ratio \( M^* \) is

\[ M^* = \left( \frac{\gamma + 1}{2} \frac{M^2}{1 + \frac{\gamma - 1}{2} M^2} \right)^{1/2} \]  

This relation is shown in figure 3 for specific-heat ratios of 1.3, 1.4, and 1.66. As the Mach number approaches infinity, \( \nu \) approaches an upper limit of \( \frac{\pi}{2} \left[ \sqrt{(\gamma + 1)/(\gamma - 1) - 1} \right] \).
Blade Design

The method of characteristics as applied to the two-dimensional isentropic flow of a perfect gas is utilized in the design of the supersonic blade sections. A description of the method of characteristics is given in references 4 and 5, and its application to the design of supersonic blade sections is given in references 2 and 6. For purposes of calculation, the flow field is considered to be divided into small regions, in each of which the flow properties are assumed to be constant. If adjacent regions are to differ slightly in properties, then the boundary between the regions must be characteristic lines and can also be shown (ref. 5) to be Mach lines. Therefore, each region is, in general, bounded either by a Mach line or a physical boundary. In figure 4 the flow field for a typical blade passage is divided by characteristic lines into a finite number of regions. The vortex-flow region is bounded by the circular arcs and the outermost vortex characteristics AE and CE. The transition arcs are composed of straight-line segments, and within each region bounded by these segments the flow is constrained to follow the wall direction. The mathematical equations necessary to define the blade are developed in the following sections.

Circular arcs. - Within the concentric circular arcs, vortex flow exists; therefore,
where \( V \) is the velocity and \( R \) is the radius in the vortex field. In this report, dimensionless parameters are used whenever possible; if this procedure is followed, equation (3) can be rewritten as

\[
\frac{V}{V_{cr}} \left( \frac{R}{r^*} \right) = \frac{\text{Constant}}{V_{cr} r^*}
\]

(4)

where \( V_{cr} \) is the critical velocity and \( r^* \) is the radius of the sonic velocity streamline in the vortex field. At \( R = r^* \), \( V = V_{cr} \); therefore, the constant is \( V_{cr} r^* \). Equation (4) then becomes

\[ M^* R^* = 1 \]

(5)

where \( M^* = V/V_{cr} \) is the dimensionless velocity and \( R^* = R/r^* \) is the dimensionless radius in the vortex field. The Prandtl-Meyer angle \( \nu \) is related to \( M^* \) through equation (1). Therefore, once \( \nu_l \) and \( \nu_u \) are specified, \( M^*_l \) and \( M^*_u \) are fixed, and the circular arc radii \( R^*_l \) and \( R^*_u \) are determined from equation (5). The amount of lower circular arc turning for the inlet and outlet portions of the blade \( \alpha_{l,i} \) and \( \alpha_{l,o} \) respectively, are

\[ \alpha_{l,i} = \beta_{l} - (\nu_{i} - \nu_{l}) \]

(6a)

and

\[ \alpha_{l,o} = \beta_{o} + (\nu_{o} - \nu_{l}) \]

(6b)

Angles measured in the counterclockwise direction are considered positive. With this convention, the inlet flow angle \( \beta_{l} \) is positive, and the outlet flow angle \( \beta_{o} \) is negative. Similarly, for the upper circular arc

\[ \alpha_{u,i} = \beta_{i} - (\nu_{i} - \nu_{l}) \]

(7a)

and

\[ \alpha_{u,o} = \beta_{o} + (\nu_{u} - \nu_{o}) \]

(7b)

The circular arcs are completely described by specification of \( \nu_{i}, \nu_{o}, \nu_{l}, \nu_{u}, \) and \( \beta_{i} \).
The outlet flow angle $\beta_0$ does not have to be specified because it can be related to $M_1$, $M_0$, and $\beta_1$ from the following consideration.

Because the inlet and outlet blade spacing is the same, the inlet and outlet blade passage areas $A_i$ and $A_o$, respectively, are related by geometry (see fig. 1) according to the equation

$$\frac{A_i}{A_o} = \frac{\cos \beta_1}{\cos \beta_0}$$

(8)

The area ratio $A_i/A_o$ can be obtained from the continuity equation (ref. 5) with the result that equation (8) becomes

$$\beta_0 = -\arccos \left( \frac{M_1}{M_0} \left( 1 + \frac{\gamma - 1}{2} \frac{M_0^2}{M_1^2} \right)^{(\gamma + 1)/(2(\gamma - 1))} \cos \beta_1 \right)$$

(9)

Lower transition arcs. - The lower surface is composed of an inlet and outlet transition arc and a circular arc (fig. 1). For symmetric blades (i.e., $\nu_i = \nu_o$), the two transition arcs are identical, and, therefore, only one needs to be calculated. For asymmetric blades, the two arcs are not identical, one being smaller (less turning) than the other. However, the smaller transition arc corresponds to a portion of the larger arc, and, consequently, only the larger arc need be calculated. If $\nu_i$ is greater than $\nu_o$, the inlet transition arc is the larger of the two arcs. For simplicity, the inlet transition arc is assumed to be the larger arc in the following discussion.

In figure 5, the lower transition arc is shown, and the nomenclature used in the computer program is indicated. The calculations are performed with respect to the nondimensional axes $x^*$ and $y^*$ (where the $x$- and $y$-coordinates are made dimensionless by dividing by $r^*$). The transition arc coordinates are generated in a sequential manner (starting at $x^* = 0$, $y^* = R_i^*$) by obtaining the intersection of the straight-line wall segments and straight Mach lines for a specified small change in flow turning. The Mach lines are determined from the outermost or major vortex-expansion characteristic, and the wall segments are determined from the flow direction. After the transition arc coordinates are calculated, they are rotated through an angle of $\alpha_{l,1}$ to obtain the coordinates of interest in the blade design (see fig. 1).

For the vortex region, it can be shown (ref. 2) that the velocity direction $\varphi$ and the dimensionless radius $R^*$ are related along the characteristic line by the equation
Figure 5. - Nomenclature used for calculation of inlet lower transition arc.
\[ \varphi = \pm \frac{1}{2} f(R^*) + \text{Constant} \quad (10a) \]

where

\[ f(R^*) = \sqrt{\frac{\gamma + 1}{\gamma - 1}} \arcsin \left( \frac{\gamma - 1}{R^*} \right) + \arcsin \left( (\gamma + 1)R^2 - \gamma \right) \quad (10b) \]

Two families of characteristics exist: the positive sign in equation (10a) gives the expansion lines, and the negative sign gives the compression lines. The major vortex-expansion-characteristic equation is

\[ \varphi = \frac{1}{2} \left[ f(R^*) - f(R_d*) \right] \quad (11) \]

since the boundary condition at \( x^* = 0 \) is that \( \varphi = 0 \) and \( R^* = R_d^* \). If the flow field is divided into small regions, the flow direction at any point along the characteristic line (as given by eq. (11)) may be considered to be equal to the flow direction within the adjacent flow region, as indicated in figure 5. For \( k \) transition arc segments, each of which produces \( \Delta \nu \) degrees of turning, the flow direction within any flow segment \( \varphi_{k,i} \) is given by

\[ \varphi_{k,i} = \nu_i - \nu_l - (k - 1)\Delta \nu = \varphi_{k+1,i} + \Delta \nu \quad (12) \]

where \( k \) is an integer that varies from 1 to \( \left[ (\nu_i - \nu_l)/\Delta \nu \right] + 1 \). At \( k = \left[ (\nu_i - \nu_l)/\Delta \nu \right] + 1 \), the flow direction is 0, and at \( k = 1 \), it is \( \nu_i - \nu_l \). Equating equations (11) and (12) and eliminating \( f(R_d^*) \) through equations (1), (5), and (10b) result in

\[ f(R_{k,i}^*) = 2\nu_i - \frac{\pi}{2} \left( \sqrt{\frac{\gamma + 1}{\gamma - 1}} - 1 \right) - 2(k - 1)\Delta \nu \quad (13) \]

which relates the dimensionless radius \( R^* \) to the incremental flow turning \( \Delta \nu \) along the major vortex-expansion characteristic. At \( k = \left[ (\nu_i - \nu_l)/\Delta \nu \right] + 1 \), \( R_{k,i}^* = R_d^* \); as \( k \) is decreased, \( R_{k,i}^* \) decreases and is obtained by the simultaneous solution of equations (10b) and (13) by an iterative procedure. Since \( R_{k,i}^* \) can be determined for any value of \( k \), the \( x^*, y^* \) coordinates along the major expansion characteristic can be obtained from
\[ x_{k,i}^* = -R_{k,i}^* \sin \phi_{k,i} \]  

and

\[ y_{k,i}^* = R_{k,i}^* \cos \phi_{k,i} \]  

These coordinates are also points on the straight Mach lines. The equation specifying the Mach line is therefore determined once the slope is obtained, which is easily accomplished because the Mach line is inclined at the Mach angle \( \mu \) to the velocity direction (see fig. 5). The Mach line, therefore, makes an angle of \( \varphi + \mu \) with respect to the \( x^* \) axis. It is possible to define the Mach line at the mean Mach angle to the mean flow direction (ref. 5) so that the slope of the Mach line \( m_{k,i} \) is given by

\[ m_{k,i} = \tan \left( \frac{\varphi_{k,i} + \varphi_{k+1, i} + \mu_{k,i} + \mu_{k+1, i}}{2} \right) \]  

where

\[ \mu_{k,i} = -\arcsin \left( \frac{1}{M_{k,i}} \right) = -\arcsin \left[ \sqrt{\left( \frac{\gamma + 1}{2} \right) \frac{R_{k,i}^*}{2} \left( \frac{\gamma - 1}{2} \right)} \right] \]  

The equation of the Mach line is therefore

\[ y^* = m_{k,i} (x^* - x_{k,i}^*) + y_{k,i}^* \]  

where \( k \) varies from 1 to \( k_{\text{max}} = (\nu_i - \nu_l)/\Delta \nu \). The equation of the transition arc segment is more easily determined since each segment is a straight line parallel to the velocity direction \( \varphi \). The slope of the wall segment \( m_{k,i} \) is then

\[ m_{k,i} = \tan \varphi_{k+1, i} \]  

and the equation for the wall segment is

\[ y^* = m_{k,i} \left[ x^* - (x_{k+1, i}^*) \right] + (y_{k+1, i}^*) \]
where \( k \) varies from 1 to \( k_{\text{max}} \) and \( x^*_l \) and \( y^*_l \) are the lower transition arc coordinates. The values of \( x^*_l \) and \( y^*_l \) are known at \( k = (v_1 - v_l)/\Delta v + 1 = k_{\text{max}} + 1 \), where \( x^*_l = 0 \) and \( y^*_l = R^*_l \). The remaining transition coordinates are generated by finding the intersection of the Mach lines with the wall segments starting at \( k = k_{\text{max}} \) and sequentially decreasing \( k \) until \( k = 1 \). The intersection of the two straight lines is given by

\[
(x^*_l)_{k,i} = \left[ \frac{(y^*_l)_{k+1,i} - \overline{m}_{k,i}(x^*_l)_{k+1,i}}{m_{k,i} - \overline{m}_{k,i}} \right] - \frac{(y^*_l)_{k,i} - m_{k,i}x^*_l)}{-m_{k,i}x^*_l} \tag{19a}
\]

and

\[
(y^*_l)_{k,i} = \frac{m_{k,i}[(y^*_l)_{k+1,i} - \overline{m}_{k,i}(x^*_l)_{k+1,i}]}{m_{k,i} - \overline{m}_{k,i}} - \frac{m_{k,i}[(y^*_l)_{k,i} - m_{k,i}x^*_l]}{-m_{k,i}x^*_l} \tag{19b}
\]

The transition arc coordinates obtained from equation (19) are rotated through an angle \( \alpha^*_{l,i} \) resulting in the \( X^* \), \( Y^* \) coordinates of interest in the blade design. The rotated coordinates \( X^*_l \) and \( Y^*_l \) are obtained from

\[
(X^*_l)_{k,i} = (x^*_l)_{k,i} \cos \alpha^*_{l,i} - (y^*_l)_{k,i} \sin \alpha^*_{l,i} \tag{20a}
\]

and

\[
(Y^*_l)_{k,i} = (x^*_l)_{k,i} \sin \alpha^*_{l,i} + (y^*_l)_{k,i} \cos \alpha^*_{l,i} \tag{20b}
\]

Upper transition arcs. - The upper surface (like the lower surface) is composed, in part, of an inlet and outlet transition arc, only one of which must be calculated. For simplicity, it is again assumed that the inlet transition arc is the larger of the two arcs. (For the upper arc, this requires that \( v_o \) be greater than \( v_l \).) The upper transition arc is shown schematically in figure 6, and the pertinent nomenclature is noted. The procedure employed to calculate the upper transition arc is analogous to that used for the lower transition arc; the resulting equations which are of similar form are not repeated herein. The subscript \( j \) is used to represent the upper transition arc coordinates where \( j \) varies from 1 to \( j_{\text{max}} = (v_u - v_l)/\Delta v \).

Geometric parameters. - After the blade calculations have been performed, a number of blade parameters of interest, including blade solidity, spacing, chord, and total
Figure 6. - Nomenclature used for calculation of inlet upper transition arc.
flow turning angle, are calculated. The blade spacing $G^*$ and chord $C^*$ are obtained from the blade coordinates, and the solidity is obtained from the ratio $C^*/G^*$. The total flow turning angle $\theta$ is obtained from the inlet and outlet flow angles. The blade coordinates are also translated by $G^*$ so that coordinates for two complete blades are obtained.

Design Limitations

The design limitations (i.e., the constraints on the choice of $\nu_l$ and $\nu_u$ for specified $\nu_i$, $\nu_o$, $\beta_i$, and $\gamma$) imposed by consideration of supersonic starting and flow separation problems have been discussed in reference 3. These limitations are calculated by the procedure described in the following paragraphs and are given as output from the computer program presented herein.

Supersonic starting. - The problem of establishing supersonic flow on startup is discussed in reference 7 for supersonic compressors. For supersonic turbines, the resulting design limitations due to starting are presented in reference 2, where it is assumed that a normal shock wave spans the blade inlet at the instant of startup. Under this condition it is necessary to ensure that the weight flow can pass through the turbine. The maximum value of the inlet Prandtl-Meyer angle ($\nu_i_{\text{max}}$) is determined by first finding the maximum weight flow through the blade passage, while taking into account the normal shock losses. This maximum weight flow is then equated to the flow rate after the shock has passed through the passage.

The weight flow through the passage is obtained by integrating the continuity equation in the vortex region

$$w = h \int_{R_u}^{R_l} \rho V \, dR$$

(21)

where $\rho$ is the density, $V$ is the velocity, and $h$ is the blade height. The density can be written as (ref. 2)

$$\rho = \rho_{i,d}' \left[ 1 - \left( \frac{\gamma - 1}{2} \right) \left( \frac{V}{a_{i,d}'} \right)^2 \right]^{1/(\gamma-1)}$$

(22)

where $\rho_{i,d}'$ and $a_{i,d}'$ are the density and the speed of sound, respectively, just downstream of the shock, and are evaluated at total conditions. For perfect gases, the total
temperature is constant through a normal shock so that \( a^_i_{,d} = a^i_1 \). Utilizing equation (3) and the definition

\[
K^* = \sqrt{\frac{\gamma - 1}{2}} \left( \frac{VR}{a^i_1 R_\ell} \right)
\]  

(23)

results in equation (21) in the following form:

\[
w = h_a^i \rho^i_1, dR_l \sqrt{\frac{2}{\gamma - 1}} \int_{R_u}^{R_l} \left( 1 - \frac{K^* R_\ell^2}{R^2} \right)^{1/(\gamma - 1)} K^* \frac{dR}{R}
\]  

(24)

Differentiating equation (24) with respect to \( K^* \) and setting the result equal to 0 give the value of \( K^* \) (denoted as \( K^*_{\text{max}} \)) for which the weight flow is a maximum. This procedure gives

\[
\frac{dw}{dK^*} = h_a^i \rho^i_1, dR_l \sqrt{\frac{2}{\gamma - 1}} \int_{R_u}^{R_l} \left[ \left( 1 - \frac{K^*_{\text{max}} R_\ell^2}{R^2} \right)^{1/(\gamma - 1)} \left( 1 - \frac{K^*_{\text{max}} R_\ell^2}{R^2} \right)^{(2-\gamma)/(\gamma - 1)} \right] \frac{dR}{R} = 0
\]  

(25)

Changing the variable from \( R \) to \( M^* \) gives, for equation (25),
Integrating the right side of equation (26) results in

\[ \int_{M^*_l}^{M^*_u} \left[ 1 - \left( \frac{K_{\text{max}}}{M^*_l} \right)^2 M^* \right] \frac{dM^*}{M^*} = \frac{2}{\gamma - 1} \int_{M^*_l}^{M^*_u} \left( \frac{K_{\text{max}}}{M^*_l} \right)^2 \left[ 1 - \left( \frac{K_{\text{max}}}{M^*_l} \right)^2 M^* \right] \frac{dM^*}{M^*} \]

(27)

Similarly, equation (24) in terms of \( M^* \) becomes

\[ w_{\text{max}} = r^* h a_i' \rho_i' d \sqrt{\frac{2}{\gamma - 1}} \int_{M^*_l}^{M^*_u} \frac{K_{\text{max}}}{M^*_l} \left[ 1 - \left( \frac{K_{\text{max}}}{M^*_l} \right)^2 M^* \right] \frac{dM^*}{M^*} \]

(28)

The value of \( K_{\text{max}}^* \) is determined from equation (27) by an iterative procedure (for given values of \( M^*_l \) and \( M^*_u \)), and then the maximum weight flow is obtained from equation (28). After the shock has passed through the turbine, the weight flow can again be obtained from equation (21), which is rewritten by substituting \( V_R = V_{\text{cr}} r^* \) and changing the variable from \( R \) to \( M^* \) to give

\[ w = r^* h \int_{M^*_l}^{M^*_u} \rho V_{\text{cr}} \frac{dM^*}{M^*} \]

(29)
Substituting the following relations (ref. 5) into equation (29)

\[
\rho = \rho_i \left(\frac{2}{\gamma + 1}\right)^{1/(\gamma - 1)} \left(\frac{\gamma + 1}{2} - \frac{\gamma - 1}{2} M^* a\right)^{1/(\gamma - 1)}
\]

(30)

and

\[
V_{cr} = a_1 \sqrt{\frac{2}{\gamma + 1}}
\]

(31)

gives

\[
w = r^* a_i \rho_i \sqrt{\frac{2}{\gamma + 1}} \left(\frac{2}{\gamma + 1}\right)^{1/(\gamma - 1)} \int_{M^*_l}^{M^*_u} \left(\frac{\gamma + 1}{2} - \frac{\gamma - 1}{2} M^* a\right)^{1/(\gamma - 1)} \frac{dM^*}{M^*}
\]

(32)

where the term multiplying \( r^* a_i \rho_i \) is defined as the weight-flow parameter. Equating the two weight flows, equations (28) and (32), results in

\[
\frac{Q}{1 - C} = \frac{p_i}{\rho_i} \frac{d}{d_i} = \frac{p_i}{\rho_i}
\]

(33)

where

\[
Q = \frac{M^*_u M^*_l}{M^*_u - M^*_l} \int_{M^*_l}^{M^*_u} \left(\frac{\gamma + 1}{2} - \frac{\gamma - 1}{2} M^* a\right)^{1/(\gamma - 1)} \frac{dM^*}{M^*}
\]

(34a)

and

\[
C = 1 - \sqrt{\frac{\gamma + 1}{\gamma - 1}} \left(\frac{\gamma + 1}{2}\right)^{1/(\gamma - 1)} \int_{M^*_l}^{M^*_u} \frac{K^*_{max}}{M^*_u - M^*_l} \left[1 - \left(\frac{K^*_{max}}{M^*_l}\right)^2 M^* a\right]^{1/(\gamma - 1)} \frac{dM^*}{M^*}
\]

(34b)
The parameter $C$ has been shown in reference 2 to be the reduction in maximum flow rate caused by two-dimensional flow. The quantity $p_i'/d/p_i'$ is the total pressure recovery for a normal shock and is given by (ref. 5)

$$
\frac{p_i'}{d/p_i'} = (M_i^*)^{2\gamma/\gamma - 1} \left[ \frac{1 - \left( \frac{\gamma - 1}{\gamma + 1} \right)(M_i^*)^2_{\text{max}}}{\left( M_i^* \right)^2_{\text{max}} - \frac{\gamma - 1}{\gamma + 1}} \right]^{1/(\gamma - 1)}
$$

(35)

The maximum inlet velocity ratio $(M_i^*)_{\text{max}}$ is obtained (for given values of $M_i^*$ and $M_u^*$) by simultaneous solution of equations (33) and (35), by using the definition in equations (34a) and (34b), and by obtaining $K^*_{\text{max}}$ from equation (27). The maximum inlet Prandtl-Meyer angle for supersonic starting $(\nu_i^*)_{\text{max}}$ is then obtained from equation (1).

Flow separation. - Analysis of supersonic blade sections (ref. 3) has shown the desirability of maintaining high surface Mach numbers to alleviate the problem of supersonic starting. Under these conditions, however, adverse pressure gradients created on the blade surfaces would be expected to cause flow separation and, consequently, poor performance. Experimental investigation of simple shapes with incompressible flow at fairly high pressure gradients (ref. 8) has indicated that if the coefficient of pressure recovery (defined as the ratio of the pressure rise to the dynamic pressure at the initial point) is less than about 1/2, flow separation may be avoided. This criterion was used in the analysis presented in reference 3 to give some indication of the design restrictions due to flow separation. For supersonic velocities the separation value of the coefficient of pressure recovery may be less than 1/2. The calculational procedure is as follows.

Flow separation can occur on both the lower and upper surfaces of the blade, but since the calculational procedure is similar for both cases only the derivation for the lower surface is presented. The flow separation criterion can be written as

$$
\frac{(p_i')_{\text{max}}}{\rho_i V_i^2} = \frac{1}{2g_c}
$$

(36)

where $(p_i')_{\text{max}}$ is the maximum lower-surface pressure possible (for given inlet conditions) without causing separation. Equation (36) can be rewritten in the form
\[
\frac{(p_t)_{\text{max}}}{p_t} = \frac{p_i}{p_i} \left[ 1 + \frac{1}{2} \frac{\rho_i V_1^2}{p_i} \right]^{1/(\gamma - 1)}
\]

(37)

where the equation has been divided by the inlet total pressure \( p_i \). Substituting the following relations (ref. 9) into equation (37)

\[
\frac{p}{p'} = \left( 1 - \frac{\gamma - 1}{\gamma + 1} M_{*}^2 \right)^{\gamma/(\gamma - 1)}
\]

(38a)

and

\[
\frac{1 - \rho V^2}{2g_c} = \frac{\gamma}{\gamma + 1} M_{*}^2
\]

\[
\frac{p}{p'} = \frac{\gamma}{\gamma + 1} M_{*}^2
\]

(38b)

and simplifying result in

\[
(M^*_\ell)_{\text{min}} = \sqrt{\frac{\gamma + 1}{\gamma - 1}} \left\{ 1 - \left[ 1 - \frac{\gamma - 1}{\gamma + 1} M_{*}^2 \right] \left[ 1 + \frac{1}{2} \left( \frac{\gamma - 1}{\gamma + 1} M_{*}^2 \right) \right]^{(\gamma - 1)/\gamma} \right\}^{1/2}
\]

(39)

Similarly, applying the same criterion to the upper surfaces gives

\[
M_{*U} = \sqrt{\frac{\gamma + 1}{\gamma - 1}} \left\{ 1 - \left[ 1 - \frac{\gamma - 1}{\gamma + 1} (M_{*U})^2_{\text{max}} \right] \left[ 1 + \frac{1}{2} \left( \frac{\gamma - 1}{\gamma + 1} (M_{*U})^2_{\text{max}} \right) \right]^{(\gamma - 1)/\gamma} \right\}^{1/2}
\]

(40)

The corresponding Prandtl-Meyer angles \((\nu_{\ell})_{\text{min}}\) and \((\nu_{U})_{\text{max}}\) are obtained from equation (1).
### TABLE I - INPUT FORMAT FOR FORTRAN COMPUTER PROGRAM

<table>
<thead>
<tr>
<th>STATEMENT</th>
<th>FORTRAN STATEMENT</th>
</tr>
</thead>
<tbody>
<tr>
<td>1 2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 21 22 23 24 25 26 27 28 29 30 31 32 33 34 35 36 37 38 39 40 41 42 43 44 45 46 47 48 49 50 51 52 53 54 55 56 57 58 59 60 61 62 63 64 65 66 67 68 69 70 71 72 73 74 75 76 77 78 79 80</td>
<td></td>
</tr>
</tbody>
</table>
DESCRIPTION OF INPUT

The input for the computer program consists of an inlet flow angle, several Prandtl-Meyer angles, the specific-heat ratio \( \gamma \), and an angular increment. In addition, three optional switches must be set which regulate the content and form of the output. All the input parameters, except \( \gamma \) and the switches, must be specified in degrees. The input variables are as follows:

- **BETAN**: inlet flow angle, \( \beta \)
- **DELV**: flow turning increment (recommended value, 0, 1), \( \Delta \nu \)
- **GAM**: specific-heat ratio, \( \gamma \)
- **IPRINT**: a value of 0 will result in printing of rotated blade coordinates \( X^* \) and \( Y^* \); a value of 1 will cause both rotated and unrotated coordinates \( X^*, Y^* \) and \( x^*, y^* \) (see fig. 1) to be printed
- **ISTART**: a value of 0 will cause both starting and blade design calculations to be printed out; a value of 1 will cause only starting calculations to be performed and printed out
- **NPlot**: a value of 0 will cause blade profile and flow passage to be plotted; a value of 1 will suppress the plot
- **VIN**: inlet Prandtl-Meyer angle, \( \nu_i \)
- **VLOW**: lower-surface Prandtl-Meyer angle, \( \nu_l \)
- **VOUT**: outlet Prandtl-Meyer angle, \( \nu_o \)
- **VUP**: upper-surface Prandtl-Meyer angle, \( \nu_u \)

The flow turning increment \( \Delta \nu \) must be specified so that \( (\nu_i - \nu_l)/\Delta \nu \), \( (\nu_o - \nu_l)/\Delta \nu \), \( (\nu_u - \nu_i)/\Delta \nu \), and \( (\nu_u - \nu_o)/\Delta \nu \) are all integers. Table I shows a sample input card.

DESCRIPTION OF OUTPUT

An example of the output obtained from the program is shown in table II. The output corresponds to the input data shown in table I and consists of tables of coordinates for the description of two blade sections, supersonic starting and flow separation parameters, and a plot of the blade profile and flow passage. This example required approximately 0.2 minute of computer running time. Each section of the output has been numbered to correspond to the following description:

1. The first output of the program is a listing of the supersonic starting parameters.
If ISTART=0, the program will continue with the blade design calculations. If ISTART=1, no further output is obtained.

(2) The next output is a listing of all the input data plus the value of the calculated outlet flow angle.

(3) If IPRINT=1, the following tables are printed:
   (a) Unrotated coordinates of the lower- and upper-surface transition arcs
   (b) Coordinates of the lower- and upper-surface circular arcs
   (c) Coordinates of the upper-surface straight-line segments
   This output is, in general, not of interest except for debugging purposes and may be omitted by setting IPRINT=0.

(4) The next output is tables of the rotated coordinates of the lower and upper transition arcs. In addition, these coordinates are translated by the value of the blade spacing so that coordinates for two blade sections are obtained. Every tenth calculation point is printed if $\Delta \nu < 0.2^\circ$; otherwise every calculation point is printed.

(5) The next output is a listing of miscellaneous parameters including
   (a) Inlet, outlet, and surface dimensionless velocities or critical velocity ratios and Mach numbers
   (b) Dimensionless blade spacing, chord and solidity
   (c) Separation limitation for the lower- and upper-surface Prandtl-Meyer angles

(6) If NPLOT=0, the final output is a printer plot of the blade profile and the flow passage. If NPLOT=1, this output is omitted.
## TABLE II - SAMPLE OUTPUT

### DESIGN OF SUPERSONIC PLACES

**Calculations for Supersonic Starting**

<table>
<thead>
<tr>
<th>Parameter</th>
<th>Value</th>
</tr>
</thead>
<tbody>
<tr>
<td>$k_{max}$</td>
<td>0.3442</td>
</tr>
<tr>
<td>$C$</td>
<td>0.0300</td>
</tr>
<tr>
<td>$q$</td>
<td>0.4701</td>
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<tr>
<td>$M_{max}$</td>
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</tr>
</tbody>
</table>

The maximum design value for $k_{max}$ is 40.000 deg, when $vel_{in}$ is 20.0000 deg, $vel_{out}$ is 59.0000 deg, $gamma$ is 3.4000.

### Design Parameters

<table>
<thead>
<tr>
<th>Parameter</th>
<th>Value</th>
</tr>
</thead>
<tbody>
<tr>
<td>$delta v$</td>
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</tr>
<tr>
<td>$vel_{in}$</td>
<td>35.0000</td>
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<td>15.0000 deg</td>
</tr>
<tr>
<td>$vel_{out}$</td>
<td>56.735 deg</td>
</tr>
</tbody>
</table>

#### LUMER SURFACE

### UNTARGET TRANSITION ARCS

<table>
<thead>
<tr>
<th>INPUT X</th>
<th>YLLUM1</th>
<th>YLUM2</th>
</tr>
</thead>
<tbody>
<tr>
<td>0</td>
<td>0.0874</td>
<td>0.6634</td>
</tr>
<tr>
<td>0.001</td>
<td>0.0874</td>
<td>0.6634</td>
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<tr>
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<td>0.6634</td>
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<tr>
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<td>0.6634</td>
</tr>
<tr>
<td>0.004</td>
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<td>0.6634</td>
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<tr>
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</table>

### TRIMMED AND TUMESCENT TRANSITION ARCS

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<tr>
<th>INPUT X</th>
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<th>YLUM2</th>
<th>YLUM3</th>
<th>YLUM4</th>
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25
TABLE II - Continued

SAMPLE OUTPUT

[Table content not legible]
### TABLE II - Continued. SAMPLE OUTPUT

#### UPPER SURFACE

<table>
<thead>
<tr>
<th>Inlet J</th>
<th>X(U)</th>
<th>Y(U)</th>
</tr>
</thead>
<tbody>
<tr>
<td>...</td>
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</table>

**UNTRIFEL TRANSITION ARCS**

<table>
<thead>
<tr>
<th>Inlet J</th>
<th>X(U)</th>
<th>Y(U)</th>
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<tbody>
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</table>

**XTRIFEL AND TRANSLATED TRANSITION ARCS**

<table>
<thead>
<tr>
<th>Inlet J</th>
<th>X(U)</th>
<th>Y(U)</th>
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</table>

<table>
<thead>
<tr>
<th>Outlet J</th>
<th>X(U)</th>
<th>Y(U)</th>
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<tr>
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</tbody>
</table>

**Notes:**
- X(U) and Y(U) represent coordinates on the upper surface.
- Outlet J coordinates are also provided.

---

**Continued:**

<table>
<thead>
<tr>
<th>Inlet J</th>
<th>X(U)</th>
<th>Y(U)</th>
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<tr>
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**Notes:**
- X(U) and Y(U) represent coordinates on the upper surface.
- Outlet J coordinates are also provided.

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**Continued:**

<table>
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<tr>
<th>Inlet J</th>
<th>X(U)</th>
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**Notes:**
- X(U) and Y(U) represent coordinates on the upper surface.
- Outlet J coordinates are also provided.

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**Continued:**

<table>
<thead>
<tr>
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**Notes:**
- X(U) and Y(U) represent coordinates on the upper surface.
- Outlet J coordinates are also provided.
<table>
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<th>X (UP)</th>
<th>Y (UP)</th>
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PROGRAM DESCRIPTION

Main Program

The main program generates a table of the inlet and outlet transition arcs of the upper, lower, and translated curves. It also computes several parameters which are pertinent to the blade description, such as Mach numbers and radii, and, as an option, plots a blade profile and flow channel. The plotting is done by subroutine PLOTMY, (ref. 10). The program variables are

- **ALPH**: fixed angle
- **ALPHLN**: rotation angle for lower-curve inlet transition arc, \( \alpha_{l,i} \)
- **ALPHLO**: rotation angle for lower-curve outlet transition arc, \( \alpha_{l,o} \)
- **ALPHUI**: rotation angle for upper-curve inlet transition arc, \( \alpha_{u,i} \)
- **ALPHUO**: rotation angle for upper-curve outlet transition arc, \( \alpha_{u,o} \)
- **ALPHUP**: temporary storage
- **ALPLOW**: temporary storage
- **ANGLE**: logical switch
- **BETAN**: see INPUT
- **BETAT**: outlet flow angle, \( \beta_o \)
- **CONVER**: conversion factor for degrees to radians
- **COSALN**: cosine of ALPHLN
- **COSALO**: cosine of ALPHLO
- **COSAUI**: cosine of ALPHUI
- **COSAUO**: cosine of ALPHUO
- **CSTAR**: blade chord, \( C^* \)
- **DALPH**: angle increment
- **DELF**: see subroutine ROOT
- **DELV**: see INPUT
- **DELXI**: 1/15 length of straight-line portion of upper-curve inlet arc
- **DELXO**: 1/15 length of straight-line portion of upper-curve outlet arc
- **EMJ**: slope of Mach lines, \( m_j \)
EMK  slope of Mach lines (eq. (15)), $m_k$
EMWJ slope of wall segments, $m_j$
EMWK slope of wall segments (eq. (17)), $m_k$
F(V, FN) internally defined function (eq. (13)), $f(R^*)$
FLO see subroutine START
FN floating point index
FOFX see subroutine ROOT
FUP see subroutine START
GAM ratio of specific heats, $\gamma$
GAMEXP $1/(\text{GAM} - 1)$
GAMM1 $(\text{GAM} - 1)/2$
GAMP1 $(\text{GAM} + 1)/2$
GRTY dummy name
GSTAR blade spacing, $G^*$
I counter
IPRINT see INPUT
ISTART see INPUT
J variable index for upper curve
JJ variable index
JDEX variable index
JMAXN number of points on upper-curve inlet transition arc
JMAXO number of points on upper-curve outlet transition arc
JMN maximum of JMAXO and JMAXN
JN variable index
JNDEX number of upper-curve transition arc points to be printed
JNN variable index
JO variable index
JOO variable index
K variable index for lower curve
KK
variable index

KKK
array required by PLOTMY (see ref. 10)

KDEX
variable index

KMAXN
number of points on lower-curve inlet transition arc

KMAXO
number of points on lower-curve outlet transition arc

KMN
maximum of KMAXN and KMAXO

KN
variable index

KNDEX
number of lower-curve transition arc points to be printed

KNN
variable index

KO
variable index

KOO
variable index

KOUNT
counter

L
counter

LLL
NP1 + NP2

LSTORE
number of points saved per 5° of turning

LSTR
temporary storage

M
counter

MAXN
integer constant

MAXO
integer constant

N
variable index

NPER
variable governing selective storage

NPLOT
see INPUT

NP1
number of points on lower curve which have been saved for plotter

NP2
number of points on translated lower curve which have been saved for plotter

NP3
number of points on upper curve which have been saved for plotter

NSUM	total number of points which have been stored for plotter

NUM
counter

P
array required by PLOTMY (see ref. 10)

PERM
\[ [(\text{GAM} + 1)/(\text{GAM} - 1)]^{1/2} \]
**PHIJ** flow direction, \( \phi_j \)

**PHIJP1** previous value of PHIJ, \( \phi_{j+1} \)

**PHIK** flow direction (eq. (12)), \( \phi_k \)

**PHIKP1** previous value of PHIK, \( \phi_{k+1} \)

**R** array for storing radii of major vortex-compression-characteristic points, \( R_j^* \)

**RA** array for storing radii of major vortex-expansion-characteristic points, \( R_k^* \)

**RECONV** conversion factor for radians to degrees

**RIN** 1/SSMIN

**RLOW** radius of circular arc of lower curve as calculated in JOKOS, \( R_l^* \)

**ROUT** 1/SSMOUT

**RUP** radius of upper-curve circular arc, \( R_u^* \)

**SAME** see subroutine START

**SIGMA** blade solidity, \( \sigma \)

**SINALN** sine of ALPHLN

**SINALO** sine of ALPHLO

**SINAUI** sine of ALPHUI

**SINAUO** sine of ALPHUO

**SM** temporary storage for Mach numbers, \( M \)

**SMIN** inlet Mach number, \( M_i \)

**SMLOW** lower-surface Mach number, \( M_{l} \)

**SMOUT** outlet Mach number, \( M_o \)

**SMS** temporary storage for velocity ratio, \( M^* \)

**SMUP** upper-surface Mach number, \( M_u \)

**SSMIN** inlet velocity ratio, \( M_{i}^* \)

**SSMLOW** lower-surface velocity ratio, \( M_{l}^* \)

**SSMOUT** outlet velocity ratio, \( M_o^* \)

**SSMUP** upper-surface velocity ratio, \( M_u^* \)

**TANBI** tangent of BETAN

**TANBO** tangent of BETAT
TEMP  temporary storage
TEMPP temporary storage
TEMPPP temporary storage
THETA total flow turning angle, θ
TR temporary storage for radii
TX temporary storage for unrotated x*-coordinates
TXLO temporary storage for values of XLOW(I)
TXUP temporary storage for values of XUP(I)
TY temporary storage for unrotated y*-coordinates
TYLO temporary storage for values of YLOW(I)
TYUP temporary storage for values of YUP(I)
UMJ Mach angle, μ_j
UMJP1 previous value of UMJ, μ_{j+1}
UMK Mach angle (eq. (15)), μ_k
UMKP1 previous value of UMK, μ_{k+1}
V temporary storage for Prandtl-Meyer angles, ν
VIMAX see subroutine START
VIN see INPUT
VLOW see INPUT
VLSPMN minimum lower-surface Prandtl-Meyer angle from separation criterion, (ν_l)_min
VNL VIN-VLOW
VOL VOUT-VLOW
VOUT see INPUT
VUI VUP-VIN
VUMAX  \frac{\pi}{2} \left( \sqrt[\gamma + 1]{\gamma - 1} - 1 \right)
VUP see INPUT
VUSPMX  maximum upper-surface Prandtl-Meyer angle from separation criterion, 
        \( \nu_u \)_{max}
VUT     VUP-VOUT
X0      see subroutine ROOT
X1      see subroutine ROOT
X2      see subroutine ROOT
XCG     \( x^* \)-coordinate of a translated-curve circular arc point
XCLOW   \( x^* \)-coordinate of a lower-curve circular arc point
XCUP    \( x^* \)-coordinate of an upper-curve circular arc point
XDOWN   array for storing \( x^* \)-coordinate of points to be plotted
XINTL   see subroutine ROOT
XLOW    array for storing \( x^* \)-coordinate of unrotated lower transition arc points, 
        \( (x_i^*)_k \)
XLOWN   array for storing \( x^* \)-coordinate of lower-curve rotated inlet transition arc 
        points, \( (X_i^*)_{k,i} \)
XLOWO   array for storing \( x^* \)-coordinate of lower-curve rotated outlet transition arc 
        points, \( (X_i^*)_{k,o} \)
XMLLOW  temporary storage for values of \(-XLOW(I)\)
XMUP    temporary storage for values of \(-XUP(I)\)
XSIN    \( x^* \)-coordinate of a point on inlet straight-line portion of upper curve
XSOUT   \( x^* \)-coordinate of a point on outlet straight-line portion of upper curve
XUP     array for storing \( x^* \)-coordinate of unrotated upper transition arc points, 
        \( (x_i^*)_j \)
XUPN    array for storing \( x^* \)-coordinate of upper-curve rotated inlet transition arc 
        points, \( (X_i^*)_{j,i} \)
XUPO    array for storing \( x^* \)-coordinate of upper-curve rotated outlet transition arc 
        points, \( (X_i^*)_{j,o} \)
YACRS1  array for storing \( y^* \)-coordinates of points for plotter
YACRS2  array for temporary storage of \( y^* \)-coordinate of translated lower-curve 
        points for plotter
YCG \hspace{.5cm} Y^*-coordinate of a translated-curve circular arc point
YCLOW \hspace{.5cm} Y^*-coordinate of a lower-curve circular arc point
YCUP \hspace{.5cm} Y^*-coordinate of an upper-curve circular arc point
YLASTI \hspace{.5cm} Y^*-coordinate of first point on inlet side of upper curve
YLASTO \hspace{.5cm} Y^*-coordinate of first point on outlet side of upper curve
YLOW \hspace{.5cm} array for storing y*-coordinate of unrotated lower transition arc points, \((y_u^*)_{k,1}\)
YLOWN \hspace{.5cm} array for storing Y*-coordinate of lower-curve rotated inlet transition arc points, \((Y_u^*)_{k,1}\)
YLOWO \hspace{.5cm} array for storing Y*-coordinate of lower-curve rotated outlet transition arc points, \((Y_u^*)_{k,0}\)
YNG \hspace{.5cm} temporary storage for Y*-coordinate of a point on translated-curve inlet transition arc
YSIN \hspace{.5cm} Y*-coordinate of a point on inlet straight-line portion of upper curve
YSNG \hspace{.5cm} Y*-coordinate of a point on inlet straight-line portion of translated upper curve
YSOUT \hspace{.5cm} Y*-coordinate of a point on outlet straight-line portion of upper curve
YSTG \hspace{.5cm} Y*-coordinate of a point on outlet straight-line portion of translated upper curve
YTG \hspace{.5cm} temporary storage for Y*-coordinate of a point on translated-curve outlet transition arc
YUP \hspace{.5cm} array for storing y*-coordinate of unrotated upper transition arc points, \((y_u^*)_{j}\)
YUPN \hspace{.5cm} array for storing Y*-coordinate of upper-curve rotated inlet transition arc points, \((Y_u^*)_{j,1}\)
YUPO \hspace{.5cm} array for storing Y*-coordinate of upper-curve rotated outlet transition arc points, \((Y_u^*)_{j,0}\)

Subroutine ROOT

Subroutine ROOT is a general routine for finding the roots of equations and is derived from the "half-interval search" method described in reference 11. This method
depends on successively halving an interval which is known to contain the desired root. Subroutine ROOT is used to calculate $R^*$ from equation (10b).

A call to ROOT has the form CALL ROOT (X0, X2, XINTL, FOFX, FUNC, X1), where the elements of the call vector are

- **X0**: lower bound of initial root interval
- **X2**: upper bound of initial root interval
- **XINTL**: initial estimate of value of root
- **FOFX**: given value of dependent variable
- **FUNC**: externally defined function
- **X1**: value of root

Other program variables are

- **A**: $\text{FOFX} - \text{F2}$
- **DELF**: convergence criterion
- **F0**: function FUNC evaluated at $XX0$
- **F2**: function FUNC evaluated at $XX2$
- **FX**: function FUNC evaluated at $X$
- **KOUNT**: count of number of iterations performed
- **X**: temporary storage for present estimate of root
- **XX0**: present value of lower bound of root interval
- **XX2**: present value of upper bound of root interval

### Subroutine START

Subroutine START is used to compute the maximum value of the inlet Prandtl-Meyer angle ($\nu_1$) for supersonic starting for various lower- and upper-surface Prandtl-Meyer angles. Several other parameters of interest are computed by START and are printed as output. Among these parameters are the vortex constant for maximum weight flow $K^*_\text{max}$, the reduction in weight flow due to two-dimensional flow $C$, and the maximum value of the inlet velocity ratio for supersonic starting ($M^*_1$).

A call to START has the form CALL START (VLOW, FLO, VUP, FUP, VIMAX), where the elements of the call vector are

- **VLOW**: lower-surface Prandtl-Meyer angle, $\nu_1$
FLO \quad \text{eq. (10b) evaluated at } R^*_v

VUP \quad \text{upper-surface Prandtl-Meyer angle, } \nu_u

FUP \quad \text{eq. (10b) evaluated at } R^*_u

VIMAX \quad \text{maximum inlet Prandtl-Meyer angle for supersonic starting, } (\nu^*_i)_{\text{max}}

Other program variables are

BINTGR \quad \text{value of integral (eq. (27b))}

C \quad \text{reduction in maximum weight flow due to two-dimensional flow (eq. (34b))}

CINTGR \quad \text{value of integral (eq. (32))}

FINL \quad \text{eq. (27) evaluated at } K^*_{\text{max}} = \text{XINTL}

F0 \quad \text{eq. (27) evaluated at } K^*_{\text{max}} = X_0

F2 \quad \text{eq. (27) evaluated at } K^*_{\text{max}} = X_2

FOFX \quad \text{eq. (27) evaluated at } K^*_{\text{max}} = \text{XAMK}

Q \quad \text{eq. (34a)}

RATIO \quad \text{ratio of } \frac{Q}{1 - C}

RLOW \quad \text{radius of circular portion of lower curve, } R^*_l

RUP \quad \text{radius of circular portion of upper curve, } R^*_u

SAME \quad \text{square of ratio of XAMK to SSMLOW}

SLOPE \quad \text{slope of line}

SSMIAK \quad \text{maximum value of entering velocity ratio for starting, } (M^*_i)_{\text{max}}

SSMLOW \quad \text{lower-surface velocity ratio, } M^*_l

SSMUP \quad \text{upper-surface velocity ratio, } M^*_u

WSTAR \quad \text{weight-flow parameter (eq. (32))}

XAMK \quad \text{vortex constant for maximum weight flow, } K^*_{\text{max}}

XINTL \quad \text{initial estimate of a parameter}

X0 \quad \text{lower bound of a parameter}

X2 \quad \text{upper bound of a parameter}

YINCPT \quad \text{y-intercept of a line}
Subroutine MSSTAR

Subroutine MSSTAR is used to determine the minimum lower-surface Prandtl-Meyer angle and the maximum upper-surface Prandtl-Meyer angle from separation considerations. This subroutine uses ROOT and ADSTR in the calculation of these angles.

A call to MSSTAR has the form CALL MSSTAR (M, N, VSSTAR), where the elements of the call vector are

M  inlet or outlet dimensionless velocity, $M_i^*$ or $M_o^*$, respectively
N  variable switch
VSSTAR  Prandtl-Meyer angle from separation criterion, $(\nu_l)_\text{min}$ or $(\nu_u)_\text{max}$

The other variables used by MSSTAR are

$$A = \frac{\pi}{4} \left( \frac{\gamma + 1}{\gamma - 1} \right)$$

$$B = \frac{1}{2} \sqrt{\frac{\gamma + 1}{\gamma - 1}}$$

$$C = \gamma - 1$$

$$D = \gamma + 1$$

MS  velocity ratio from separation criterion, $(M_i^*)_{\text{min}}$ or $(M_o^*)_{\text{max}}$
X0  see subroutine ROOT
X2  see subroutine ROOT
XINTL see subroutine ROOT
FOFX see subroutine ROOT
SQRDMS $(MS)^2$

Subroutine SIMPS1

This function subprogram is used to perform numerical integration of explicit functions of one variable. The integration is performed by a modification of Simpson's rule, in which a sufficient number of intervals is used to assure six or more significant figures in the result.

A call to SIMPS1 has the form $\text{ANSWER} = \text{SIMPS1} (\text{XMIN, XMAX, FUNC1, KER})$, where the elements of the call vector are
XMIN  lower limit of integration
XMAX  upper limit of integration
FUNC1 externally defined function of a single variable
KER   storage for flagging result if necessary
Other program variables are
A     array for storing functional values at certain partition points
ANS   sum of subapproximations
B     array for storing functional values at certain partition points
C     array for storing functional values at certain partition points
E     array for storing difference terms
FRAC  variable tolerance used for subapproximations
H     distance between successive points of partition
K     variable index
N     counter
NE    equivalent to E
NTEST equivalent to TEST
P     array for storing successive subapproximations
Q     sum of difference terms
SIMPS1 value of desired integral
TEST  tester for subapproximations
T     tolerance for difference terms
V     array for storing partition points of interval

Function Subprograms

The following function subprograms are used intermittently throughout the main program and subroutines:

FUNCTION ALFUNC (A, B, Y), defined by ALFUNC where

\[ ALFUNC = \frac{1}{Y} \left( A - BY^2 \right)^{1/(\gamma - 1)} \]
FUNCTION CFACT (Y), defined by CFACT where

\[
\text{CFACT} = \frac{1}{Y} \left[ 1 - \left( \frac{\frac{K_{\text{max}}}{M^*}}{M^*} \right)^2 \right]^{1/(\gamma - 1)}
\]

FUNCTION QFACT (Y), defined by QFACT where

\[
\text{QFACT} = \frac{1}{Y} \left( \frac{\gamma + 1}{2} - \frac{\gamma - 1}{2} \right) \left( \frac{\gamma + 1}{2} \right)^{1/(\gamma - 1)}
\]

FUNCTION FRAT (Y), defined by FRAT where

\[
\text{FRAT} = \frac{2\gamma}{Y^{\gamma - 1}} \left( \frac{\gamma + 1}{2} \right)^{1/(\gamma - 1)}
\]

FUNCTION FOFRS (X), defined by FOFRS where

\[
\text{FOFRS} = \sqrt{\frac{\gamma + 1}{\gamma - 1}} \arcsin \left( \frac{\gamma - 1}{X^2} \right) + \arcsin \left( (\gamma + 1)X^2 - \gamma \right)
\]

FUNCTION FKMAX (Y, L), defined by FKMAX where

\[
\text{FKMAX} = \int_{M^*_l}^{M^*_u} \left[ 1 - \left( \frac{Y}{M^*_l} \right)^2 \right]^{1/(\gamma - 1)} \frac{dZ}{Z} + \left[ 1 - \frac{Y^2 \left( \frac{M^*_u}{M^*_l} \right)^2}{\frac{M^*_u}{M^*_l}} \right]^{1/(\gamma - 1)} - \left( 1 - Y^2 \right)^{1/(\gamma - 1)}
\]

and \( L \) is an optional switch.

FUNCTION ADSTR (X), defined by ADSTR where

\[
\text{ADSTR} = \sqrt{\frac{\gamma + 1}{\gamma - 1}} \left\{ 1 - \left[ 1 - \left( \frac{\gamma - 1}{\gamma + 1} \right)X^2 \right] \left[ 1 + \frac{1}{2} \left( \frac{\gamma - 1}{\gamma + 1} \right)X^2 \right] \right\}^{(\gamma - 1)/\gamma} \]

\[
1/2
\]
PROGRAM LISTING

COMMON/EXPALF/GAMEXP
COMMON/ROots/DELv
COMMON/FACTOR/PERM,SAME,GAM,GAMM1,GAMPI,SSMLow,SSMUp,RECONv,GRTY

DIMENSION R(800),RA(800),XLOW(800),YLOW(800),XLOWN(800),YLOWN(800)
1,XUP(800),YUP(800),XLOWO(800),YLOWO(800),XUPN(800),YUPN(800),
2XUPO(800),YUPO(800)
DIMENSION XDOWN(400),YACRS1(400),YACRS2(200),KKK(14),P(20)

LOGICAL ANGLE

EXTERNAL FOFRS

F(V,FN) = (2.*V) - ((3.14159265/2.)*PERM) - (2.*FN*DELv)

INPUT AND TITLE

ISTART=O FOR STARTING AND DESIGN ISTART=1 FOR STARTING ONLY

NPLOT=0 IF PLOT IS DESIGNED NPLOT=1 IF PLOT IS NOT DESIRED

IPRINT=0 PRINT ROTATED COORDINATES ONLY IPRINT=1 PRINT UNROTATED

AND ROTATED COORDINATES

1 READ (5,11) VIN,VOUT,BETAN,VLOW,VUP,DELV,GAM,ISTART,NPLOT,IPRINT
11 FORMAT (7(F6.2,2X),3(I1,2X))

WRITE (6,99)
99 FORMAT (IHI,3FJX,53HO E S I GN OF S U P E R S O N I C B L A


CONVERSION FACTORS AND CONSTANTS

CONVER = .174532925E-01
RECONV = 57.2957796

ONE POINT WILL BE PRINTED FOR EVERY NPER POINTS CALCULATED

IF (DELv .GE. 0.2) GO TO 12
NPER = 10
GO TO 13

12 NPER = 1

13 GAMP1 = (GAM + 1.)/2.
GAMM1 = (GAM-1.)/2.
GAMEXP = 1./(GAM-1.)
PERM = SQRT(GAMM1/GAMM1)
DELv = 0.000001
X0 = 1./PERM
X2 = 0.99999999
XINTL = (X0 + X2)/2.
LSTOR = (5.0/DELv)/FLOAT(NPER)
Dalph = 1.0*CONVER

ANGLE = .TRUE.
IF (VLOW .LE. AMIN(VUP,VIN,VOUT)) GO TO 120
WRITE (6,119)
119 FORMAT (//3IX,70HV(LOW) MUST BE LESS THAN OR EQUAL TO THE MINIMUM
10 OF V(UP),V(IN),V(OUT))
    ANGLE = .FALSE.

120 IF (VUP .GE. AMAX1(VIN,VOUT)) GO TO 118
    WRITE (6,117)
117 FORMAT (//33X,66HV(UP) MUST BE GREATER THAN OR EQUAL TO THE MAXIMU
1M OF V(IN),V(OUT))
    ANGLE = .FALSE.

118 VUMAX = (3.14159265/2.)*(PERM-1.)*RECONV
    IF (VUP .LE. VUMAX) GO TO 116
    WRITE (6,115) VUMAX
115 FORMAT (//41X,37HV(UP) MUST BE LESS THAN V(UP)(MAX) = ,F9.4,4H DEG
1)
    ANGLE = .FALSE.

116 IF (.NOT. ANGLE) GO TO 1

CC PARAMETERS FOR STARTING

    VLOW = VLOW*CONVER
    FLO = F(VLOW,1.0)
    VUP = VUP*CONVER
    FUP = F(VUP,1.0)
    CALL START (VLOW,FLO,VUP,FUP,VIMAX)

    IF (ISTART .NE. 0) GO TO 1
    WRITE (6,97)
97 FORMAT (//5X,17HDESIGN PARAMETERS)

CC MISCELLANEOUS CALCULATIONS

    DELV = DELV*CONVER
    FN = 1.
    V = VIN*CONVER
    DO 4 I=1,2
        FOFX = F(V,FN)
        CALL ROOT (X0,X2,XINTL,FOFX,FOFRS,X1)
        IF (I .EQ. 2) GO TO 4
        RIN = XI
    V = VOUT*CONVER
4 CONTINUE

    ROUT = XI

    SSMIN = 1./RIN
    CALL MSSTAR (SSMIN,0,VLSPMN)
    SSMOUT = 1./ROUT
    CALL MSSTAR (SSMOUT,1,VUSPMX)

    SMS = SSMIN
    I = 1
16 SM = SORT((((1./GAMPI)*SMS*SMS)/(1.-((GAMML1/GAMPI)*SMS*SMS))))
    GO TO (17,18,19,20),I
17 SMIN = SM
    SMS = SSMOUT
CC I = 2
GO TO 16

18 SMOUT = SM
   TEMP = (((GAMMI*SMOUT*SMOUT)+1.)/((GAMMI*SMIN*SMIN)+1.))*(GAMP1
   1/(2.*GAMMI))
   BETAN = BETAN*CONVER
   BETAT = -ARCOS(COS(BETAN)*(SMIN/SMOUT)*TEMP)
   BETAN = BETAN*RECONV
   DELV = DELV*RECONV

CC PRINT ALL DESIGN PARAMETERS
   WRITE (6,95) BETAN,VIN,VUP,VOUT,BETAT
95 FORMAT (/2X,11H BETA(IN) = ,F7.4,4H OEG,4X,8HV(IN) = ,F7.4,4H OEG,
      16X,8HV(UP) = ,F8.4,4H OEG,7X,9HV(OUT) = ,F7.4,4H OEG,4X,12HBETA(OU
      2T) = ,F8.4,4H OEG)
   WRITE (6,94) DELV, VLOW, GAM
94 FORMAT (/20X,10HDELTA V = ,F7.4,4H DEG,11X,9HV(LOW) = ,F7.4,4H DEG
      1,11X,8HGAMMA = ,F7.4)

CC CONVERT FROM DEGREES TO RADIANS
   VIN = VIN*CONVER
   VOUT = VOUT*CONVER
   VUP = VUP*CONVER
   VLOW = VLOW*CONVER
   BETAN = BETAN*CONVER
   BETAT = BETAT*CONVER
   DELV = DELV*CONVER

CC CHOOSE LONGEST TRANSITION ARC OF LOWER SURFACE
   VNL = VIN - VLOW
   KMAXN = (VNL/DELV) + 0.5
   VOL = VOUT - VLOW
   KMAXO = (VOL/DELV) + 0.5
   KMN = MAXO(KMAXN,KMAXO)
   V = AMAXI(VIN,VOUT)

CC CALCULATE R*(LOW):RLOW, M*(LOW)=SSMLOW, M(LOW)=SSMLOW
   IF (VLOW .EQ. 0.0) GO TO 2
   FN = KMN + 1
   FOFX = F(V,FN)
   CALL ROOT (XO,X2,XINTL,FOFX,FOFRS,RLOW)
   GO TO 3
2 RLOW = 1.0
3 SSMLOW = 1./RLOW
   SMS = SSMLOW
   I = 3
   GO TO 16
19 SMLOW = SM

CC SET INITIAL POINTS FOR LOWER ARC CALCULATIONS
   KNDEX = KMN/NPER
   KDEX = KNDEX
   RA(KDEX+1) = RLOW
   XLOW(KDEX+1) = 0.0
   YLOW(KDEX+1) = RLOW
PHIKPI = -(V-VLOW) + FLOAT(KMN)*DELV
UMKPI = ARSIN(SQRT(GAMPI*RLOW*RLOW - GAMMI))
TXLO = XLOW(KDEX+1)
TYLO = YLOW(KDEX+1)
ALPHLN = VNL - BETAN
ALPHLO = -(VOL+BETAT)

IF (ALPHLN .LE. 0.0 .AND. ALPHLO .GE. 0.0) GO TO 110
ANGLE = .FALSE.
WRITE (6,111)

111 FORMAT (//27X,79HV(LOW) MUST BE GREATER THAN OR EQUAL TO V(IN) - BIETA(IN) ANO V(OUT) + BIETA(OUT))

CC CHOOSE LONGEST TRANSITION ARC OF UPPER SURFACE

110 VUT = VUP - VOUT
JMAXO = (VUT/DELV)+0.5
VUI = VUP - VIN
JMAXN = (VUI/DELV)+0.5
JMN = MAXO(JMAXO,JMAXN)
V = AMINI(VOIUT,VIN)

CC CALCULATE R_(UP):RUP, M*(UP)=SSMUP, M(UP)=MUP
FN = -(JMN+1) + 2
FOFX = F(V,FN)
CALL ROOT (X0,X2,XINTL,FOFX,FOFRS,ROP)
SSMUP = ./RUP
SMS = SSMUP
I = 4
GO TO 16

20 SMUP = SM

CC SET INITIAL POINTS FOR UPPER ARC CALCULATIONS
JNDEX = JMN/NPER
JDEX = JNDEX
R(JDEX+1) = RUP
XUP(JDEX+1) = 0.0
YUP(JDEX+1) = RUP
PHIJPI = -(VUP-V) + FLOAT(JMN)*DELV
UMJPI = ARSIN(SQRT(GAMPI*RUP*RUP - GAMMI))
TXUP = XUP(JDDEX+I)
TYUP = YIJP(JDEX+I)
ALPHU[ = VUI - BETAN
ALPHUO = -(VUT+BETAT)

IF (ALPHUI .LE. 0.0 .AND. ALPHUO .GE. 0.0) GO TO 112
ANGLE = .FALSE.
WRITE (6,113)

113 FORMAT (//28X,75HV(UP) MUST BE LESS THAN OR EQUAL TO V(IN) + BIETA(IN) ANO V(OUT) - BIETA(OUT))

112 IF (.NOT. ANGLE) GO TO 1

IF (VIN .EQ. VLOW .AND. VLOW .EO. VOUT) GO TO IO0

C****CALCULATE COORDINATES FOR LOWER TRANSITION ARC - UNROTATED
KDEX = KNDEX + 1
NUM = 0
V = AMAX1(VIN, VOUT)
DO 30 KK = 1, KMN
K = (KM N + 1) - KK
NUM = NUM + 1
PHIK = PHIKP1 - DELV
FN = K
FOFX = F(V, FN)
CALL ROOT (X0, X2, XINTL, FOFX, FOFRS, TR)
TX = TR*S IN (PHIK)
TY = TR*COS(PHIK)
EMWK = TAN(-PHIKP1)
UMK = ARSIN(SORT(AMIND*TR - TR*TRM1))
EMK = -TAN((PHI + UMK + PHIKP1 + UMKP1)/2.)
TEMP = TYLO - EMWK*TXLO
TEMPP = TY - EMK*TX
TEMPPP = EMK - EMWK
TXLO = (TEMP - TEMPP)/TEMPPP
TYLO = (((EMK = TEMP) - (EMWK*TEMPP))/TEMPPP
PHIKP1 = PHIK
UMKP1 = UMK

CC SAVE EVERY =NPER-TH= POINT
N = NUM - (NUM/NPER) NPER
IF (N.GT.0) GO TO 30
KDEX = KDEX - 1
RA(KDEX) = TR
XLOW(KDEX) = TXLO
YLOW(KDEX) = TYLO
30 CONTINUE

C****CALCULATE COORDINATES FOR LOWER TRANSITION ARC - ROTATED
100 KDEX = KNDEX + 1
KMN = KM N + 1
SINALN = SIN(ALPHLN)
COSALN = COS(ALPHLN)
SINALO = SIN(ALPHLO)
COSALO = COS(ALPHLO)
KN = (KMAXN/NPER) + 2
KO = (KMAXO/NPER) + 2
DO 40 KK = 1, KDEX
K = (KDEX+1) - KK
KN = KN - 1
KO = KO - 1
IF (KN.LE.0) GO TO 401
XLOWN(KN) = YLOW(K)*SINALN + XLOW(K)*COSALN
YLOWN(KN) = YLOW(K)*COSALN - XLOW(K)*SINALN
401 IF (KO.LE.0) GO TO 40
XLOWO(KO) = YLOW(K)*SINALO - XLOW(K)*COSALO
YLOWO(KO) = YLOW(K)*COSALO + XLOW(K)*SINALO
40 CONTINUE

IF (VIN.EQ. VUP .AND. VUP.EQ. VOUT) GO TO 200

C****CALCULATE COORDINATES FOR UPPER TRANSITION ARC - UNROTATED
CC
JDEX = JNDEX + 1
NUM = 0
V = AMIN(VOUT,VIN)
DO 41 JJ=1,JMN
J = (JMN+1) - JJ
NUM = NUM + 1
PHIJ = PHIJP1 - DELV
FN = -J + 2
FOFX = F(V,FN)
CALL ROOT (X0,X2,XINTL,FOFX,FOFRS,TR)
TX = TR*SIN(PHIJ)
TY = TR*COS(PHIJ)
EMWJ = TAN(-PHIJP1)
UMJ = ARSIN(SQRT(GAMPI*TR*TR - GAMMI))
EMJ = TAN((-PHIJ+UMJ-PHIJP1+UMJP1)/2.)
TEMP = TYUP - EMWJ*TXUP
TEMPP = TY - EMJ*TX
TEMPPP = EMJ - EMWJ
TXUP = (TEMP - TEMPP)/TEMPPP
TYUP = ((EMJ*TEMP) - (EMWJ*TEMPP))/TEMPPP
PHIJP1 = PHIJ
UMJP1 = UMJ
CC
SAVE EVERY =NPER-TH= POINT
N = NUM - (JMN/NPER)*NPER
IF (N .GT. 0) GO TO 41
JDEX = JDEX - 1
R(JDEX) = TR
XUP(JDEX) = TXUP
YUP(JDEX) = TYUP
41 CONTINUE
C****CALCULATE COORDINATES FOR UPPER TRANSITION ARC - ROTATED
200 JDEX = JNDEX + 1
JMN = JMN + 1
SINAUI = SIN(ALPHUI)
COSAUI = COS(ALPHUI)
SINAUO = SIN(ALPHUO)
COSAUD = COS(ALPHUO)
JN = (JMAXN/NPER) + 2
JO = (JMAXO/NPER) + 2
DO 47 JJ=1,JDEX
J = (JDEX+1) - JJ
JO = JO - 1
JN = JN - 1
IF (JO .LE. 0) GO TO 471
XUPO(JO) = YUP(J)*SINAUO - XUP(J)*COSAUI
YUPO(JO) = YUP(J)*COSAUI + XUP(J)*SINAUO
471 IF (JN .LE. 0) GO TO 47
XUPN(JN) = YUP(J)*SINAU + XUP(J)*COSAUI
YUPN(JN) = YUP(J)*COSAUI - XUP(J)*SINAU
47 CONTINUE
CC
CALCULATE G* - THE DIMENSIONLESS BLADE SPACING
TANBI = TAN(BETAN)
YLASTI = YUPN(1) + TANBI*(XLOWN(1) - XUPN(1))

47
GSTAR = YLOWN(1) - YLAST

CC TITLES
WRITE (6,93)
93 FORMAT (54X,25HL O W E R S U R F A C E)
IF (IPRINT .EQ. 0) GO TO 844
WRITE (6,88)
88 FORMAT (54X,25HUNROTATED TRANSITION ARCS)
WRITE (6,87)
87 FORMAT (54X,25HINLET K,4X,8HX*(LOW) ,3X,8HY*(LOW) ,68X,8HY*(LOW)
1,3X,8HX*(LOW) ,2X,8HOUTLET K)

C****PRINT COORDINATES FOR LOWER TRANSITION ARC - UNROTATED
KDEX = KINDEX + 2
DO 51 KK=1,KMN,NPER
  K = (KMN+1) - KK
  KN = (KMAXN+2) - KK
  KO = (KMAXO+2) - KK
  KDEX = KDEX - 1
  XMLOW = -XLOW(KDEX)
  IF (KN .GT. 0 .AND. KO .GT. 0) GO TO 510
  IF (KN .LE. 0) GO TO 511
  WRITE (6,861) KN,XLOW(KOEX),YLOW(KI)EX)
      861 FORMAT (4X,14,SX,F8,4,3X,F8.A)
  GO TO 51
  511 WRITE (6,863) YLOW(KDEX),XMLOW,KO
      863 FORMAT (IOOX,FB.4,3X,FR.4,SX,14)
  GO TO 51
  510 WRITE (6,860) KN,XLOW(KDEX),YLOW(KDEX),YLOW(KDEX),XMLOW,KO
      860 FORMAT (4X,I4,5X,F8.4,3X,F8.4,5X,I4)
  51 CONTINUE

CC TITLES
844 WRITE (6,92)
92 FORMAT (54X,25HROTTATED AND TRANSLATED TRANSITION ARCS)
WRITE (6,84)
84 FORMAT (54X,25HINLET K,4X,8HX*(LOW) ,3X,8HY*(LOW) ,68X,8HY*(LOW)
1,3X,8HX*(LOW) ,2X,8HOUTLET K)

C****PRINT COORDINATES FOR LOWER TRANSITION ARC - ROTATED
M = 1
  XDOWN(M) = XLOWN(1)
  YACRS1(M) = YLOWN(1)
  YACRS2(M) = YLOWN(1) - GSTAR
  M = M + 1
  XDOWN(M) = XLOWO(1)
  YACRS1(M) = YLOWO(1)
  YACRS2(M) = YLOWO(1) - GSTAR

CC STORE POINTS FOR PLOTTING - ONE POINT FOR EVERY FIVE DEGREES OF TURNING
  MAXN = (KMAXN/NPER) + 1
  MAXO = (KMAXO/NPER) + 1
  KNN = MAXN + 1
  KOO = MAXO + 1
  I = 0
DO 55 KK=1,KN,NPER
KN = (KMAXN+2) - KK
KO = (KMAXO+2) - KK
KNN = KNN - 1
KOO = KOO - 1
I = I + 1
LSTR = LSTORE=I
IF (KN .GT. 0 .AND. KO .GT. 0) GO TO 550
IF (KN .LE. 0) GO TO 551
IF (LSTR .GT. MAXN) GO TO 559
M = M + 1
XDOWN(M) = XLOWN(LSTR)
YACRS1(M) = YLOWN(LSTR)
YACRS2(M) = YLOWN(LSTR) - GSTAR
559 YNG = YLOWN(KNN) - GSTAR
WRITE (6,831) KN,XLOWN(KNN),YLOWN(KNN),YNG
831 FORMAT (4X,I4,7X,F8.4,6X,F8.4,6X,F8.4)
GO TO 55
551 IF (LSTR .GT. MAXO) GO TO 557
M = M + 1
XDOWN(M) = XLOWO(LSTR)
YACRS1(M) = YLOWO(LSTR)
YACRS2(M) = YLOWO(LSTR) - GSTAR
557 YTG = YLOWO(KOO) - GSTAR
WRITE (6,833) YTG,YLOWO(KOO),XLOWO(KOO),KO
833 FORMAT (4X,I4,7X,F8.4,6X,F8.4,6X,F8.4,7X,I4)
GO TO 55
550 YNG = YLOWN(KNN) - GSTAR
YTG = YLOWO(KOO) - GSTAR
IF (LSTR .GT. MAXN) GO TO 558
M = M + 1
XDOWN(M) = XLOWN(LSTR)
YACRS1(M) = YLOWN(LSTR)
YACRS2(M) = YLOWN(LSTR) - GSTAR
558 IF (LSTR .GT. MAXO) GO TO 556
M = M + 1
XDOWN(M) = XLOWO(LSTR)
YACRS1(M) = YLOWO(LSTR)
YACRS2(M) = YLOWO(LSTR) - GSTAR
556 WRITE (6,830) KN,XLOWN(KNN),YLOWN(KNN),YNG,YTG,YLOWO(KOO),XLOWO(KO),KO
830 FORMAT (4X,I4,7X,F8.4,6X,F8.4,6X,F8.4,30X,F8.4,6X,F8.4,6X,F8.4,7X,114)
55 CONTINUE

M = M + 1
XDOWN(M) = XLOWN(MAXN)
YACRS1(M) = YLOWN(MAXN)
YACRS2(M) = YLOWN(MAXN) - GSTAR
M = M + 1
XDOWN(M) = XLOWO(MAXO)
YACRS1(M) = YLOWO(MAXO)
YACRS2(M) = YLOWO(MAXO) - GSTAR
C***CIRCULAR ARC (LOWER)
IF (IPRINT .EQ. 0) GO TO R10
WRITE (6,81)
81 FORMAT (/60X,13HCIRCULAR ARCS//40X,8HX*C(LOW),3X,8HY*C(LOW),16X,
1 8HX*C(LOW),3X,11HY*C(LOW)-G*)

810 M = M + 1
    XDOWN(M) = 0.0
    YACRS1(M) = RLOW
    YACRS2(M) = RLOW - GSTAR
    THETA = (BETAN - BETAT)*RECONV
    ALPH = ALPHLO + DALPH
    ALPLOW = ALPHLN
    KOUNT = 0
60 XCLOW = RLOW*SIN(ALPLOW)
    YCLOW = RLOW*COS(ALPLOW)
    XCG = XCLOW
    YCG = YCLOW - GSTAR
    KOUNT = KOUNT + 1

IF (KOUNT .NE. LSTORE) GO TO 601
KOUNT = 0
M = M + 1
    XDOWN(M) = XCLOW
    YACRS1(M) = YCLOW
    YACRS2(M) = YCG

601 IF (IPRINT .EQ. 0) GO TO 800
    WRITE (6,80) XCLOW,YCLOW,XCG,YCG
80 FORMAT (39X,FB.4,3X,FB.4,16X,FS°4,3X,F8.4)
800 ALPLOW = ALPLOW + DALPH
    IF (ABS(ALPH-ALPLOW) .LT. 0.001) GO TO 56
    IF (ALPHLO .LT. ALPLOW .AND. ALPLOW .LT. ALPH) ALPLOW = ALPHLO
    GO TO 60

CC STORE THE TRANSLATED LOWER ARC FOR PLOTTER
56 NPI = M
    DO 2000 I=1,NPI
    M = M + 1
    XDOWN(M) = XDOWN(I)
2000 YACRS1(M) = YACRS2(I)
    NP2 = NP1

CC TITLES
WRITE (6,79)
79 FORMAT (1HI,53X,25HU P P E R S U R F A C E)
    IF (IPRINT .EQ. 0) GO TO 700
    WRITE (6,74)
74 FORMAT (/54X,25HUNROTATED TRANSITION ARCS)
WRITE (6,73)
73 FORMAT (/32X,8H INLET J,3X,HH X*(UP),3X,8H Y*(UP),68X,8H Y*(UP)
1,3X,8H X*(UP),3X,8HOUTLET J)

C***PRINT COORDINATES FOR UPPER TRANSITION ARC - UNROTATED
JDEX = JINDEX + 2
    DO 65 JJ=1,JMN,NPER
    J = (JMN+1) - JJ
    JO = (JMAXO+2) - JJ

50
JN = (JMAXN+2) - JJ
JDEX = JDEX - 1
XMUP = -XUP(JDEX)
IF (JN .GT. 0 .AND. JO .GT. 0) GO TO 650
IF (JN .LE. 0) GO TO 651
WRITE (6,720) JN,XUP(JDEX),YUP(JDEX)
    720 FORMAT (4X,14,5X,F8.4,3X,F8.4)
    GO TO 65
650 WRITE (6,723) YUP(JDEX),XMUP,JO
    723 FORMAT (100X,F8.4,3X,F8.4,5X,14)
    GO TO 65
651 WRITE (6,721) JN,XUP(JDEX),YUP(JDEX),XMUP,JO
    721 FORMAT (4X,14,5X,F8.4,3X,F8.4,6X,F8.4,3X,F8.4,5X,14)
    65 CONTINUE

CC TITLES
700 WRITE (6,78)
    78 FORMAT (IHL,46X,38HROTATED AND TRANSLATED TRANSITION ARCS)
WRITE (6,70)
    70 FORMAT (//2X,SH INLET J,6X,TH X*(UP),TX,TH Y*(UP),6X,9HY*(UP)+G*,
129X,9HY*(UP)+G*,6X,7H Y*(UP),?X,TH X*(UP),SX,8HOUTLET J)
    L = L + 1
C****PRINT COORDINATES FOR UPPER TRANSITION ARC - ROTATED
CC STORE POINTS FOR PLOTTER - ONE POINT FOR EVERY FIVE DEGREES OF TURNING
L = L + 1
XDOWN(L) = XUPN(JN)
YACRS1(L) = YUPN(JN)
L = L + 1
XDOWN(L) = XUPO(JN)
YACRS1(L) = YUPO(JN)
MAXO = (JMAXO/NPER) + 1
MAXN = (JMAXN/NPER) + 1
JO0 = MAXO + 1
JNN = MAXN + 1
I = 0
DO 303 JJ = 1,JMN,NPER
    JO = (JMAXO+2) - JJ
    JN = (JMAXN+2) - JJ
    JOO = JO0 - 1
    JNN = JNN - 1
    I = I + 1
    LSTR = LSTORE*I
    IF (JN .GT. 0 .AND. JO .GT. 0) GO TO 3030
    IF (JN .LE. 0) GO TO 3031
    IF (LSTR .GT. MAXN) GO TO 688
    L = L + 1
    XDOWN(L) = XUPN(LSTR)
    YACRS1(L) = YUPN(LSTR)
688 YNG = YUPN(JNN) + GSTAR
WRITE (6,68) JN,XUPN(JNN),YUPN(JNN),YNG
    68 FORMAT (4X,14,7X,F8.4,6X,F8.4,6X,F8.4)
    GO TO 303
3031 IF (LSTR .GT. MAXO) GO TO 689
L = L + 1
XDOWN(L) = XUP0(LSTR)
YACRS1(L) = YUPO(LSTR)

689 YTG = YUPO(J00) + GSTAR
WRITE (6,683) YTG,YUPO(J00),XUPO(J00),J0
683 FORMAT (81X,F8.4,6X,F8.4,6X,F8.4,7X,14)
GO TO 303

3030 YNG = YUPN(JNN) + GSTAR
YTG = YUPO(J00) + GSTAR
IF (LSTR .GT. MAXN) GO TO 670
L = L + 1
XDOWN(L) = XUPN(LSTR)
YACRS1(L) = YUPN(LSTR)

670 IF (LSTR .GT. MAXD) GO TO 671
L = L + 1
XDOWN(L) = XUP0(LSTR)
YACRS1(L) = YUPO(LSTR)

671 WRITE (6,680) JN,XUPN(JNN),YUPN(JNN),YNG,YTG,YUPO(J00),XUPO(J00),
106 XDOWN(L) = XUPN(MAXN)
YACRS1(L) = YUPN(MAXN)
L = L + 1
XDOWN(L) = XUPO(MAXO)
YACRS1(L) = YUPO(MAXO)

C****CIRCULAR ARC (UPPER)
IF (IPRINT .EQ. O) GO TO 6700
WRITE (6,67)
67 FORMAT (//,60X,13HCIRCULAR ARCS//4OX,8HX*C(UP) ,3X,SHY*C(UP) ,16X,
1 8HX*C(UP) ,3X,10HY*C(UP)+G*)

6700 L = L + 1
XDOWN(L) = 0.0
YACRS1(L) = RUP

ALPH = ALPHUO + DALPH
ALPHUP = ALPHUI
KOUNT = 0
305 XCUP = RUP*SIN(ALPHUP)
YACRS1(L) = YCUP + GSTAR
KOUNT = KOUNT + 1
IF (KOUNT .NE. LSTORE) GO TO 672
KOUNT = 0
L = L + 1
XDOWN(L) = XCUP
YACRS1(L) = YCUP

672 IF (IPRINT .EQ. O) GO TO 660
C****CALCULATE COORDINATES FOR STRAIGHT LINE PORTION OF UPPER ARC
CC FIFTEEN POINTS ARE CALCULATED FOR PLOTTING PURPOSES
306 IF (IPRINT .EQ. 0) GO TO 3070
WRITE (6,307)
307 FORMAT ('//59X,14HSTRAIGHT LINES//5X,8H X*S(IN),5X,8H Y*S(IN) ,3X, 
110HY*S(IN)+G*,54X,11HY*S(OUT)+G*,2X,8HY*S(OUT),5X,8HX*S(OUT))
3070 KOUNT = -1
DELX1 = ( XUPN(1) - XLOWN(1) )/15.
DELX0 = ( XLOWO(1) - XUPO(1) )/15.
XSIN = XUPN(1)
YSIN = YUPN(1)
XSOUT = XUPO(1)
YSOUT = YUPO(1)
TANBO = TAN(BETAT)
GO TO 309
310 XSIN = XSIN - DELX1
XSOUT = XSOUT + DELX0
YSIN = YUPN(1) + TANBI*(XSIN - XUPN(1))
YSOUT = YUPO(1) + TANBO*(XSOUT - XUPO(1))
309 YSNG = YSIN + GSTAR
YSTG = YSOUT + GSTAR
313 FORMAT (5X,FR.4,4X,FR.4,4X,FR.4,4X,FR.4)
3133 IF (XSOUT .GE. XLOWO(1) ) GO TO 310
IF (N *.GT. 0) GO TO 673
L = L + 1
XDOWN(L) = XSIN
YACRSI(L) = YSIN
673 IF (IPRINT .EQ. 0) GO TO 3133
WRITE (6,313) XSIN,YSIN,YSNG
3133 FORMAT (5X,FR.4,4X,FR.4,4X,FR.4)
GO TO 310
312 NP3 = L - (NP1 + NP2)
NSUM = NP1 + NP2 + NP3 + 1
XDOWN(1NSUM) = 0.0
YACRS1(INSUM) = 0.0

C****MISCELLANEOUS CALCULATIONS
WRITE (6,622)

622 FORMAT (/54X,24HMISCELLANEOUS PARAMETERS//)
YLASTO = YUPO(1) + TANBO*(XLOWO(1) - XUPO(1))
CSTAR = SQRT(((XLOWO(1) - XLOW(1))**2) + ((YLOWO(1) - YLOW(1))
1**2))
SIGMA = CSTAR /GSTAR
WRITE (6,999) VLSPMN, VUSPMX

999 FORMAT (17X,84HTHE MINIMUM LOWER SURFACE PRANDTL-MEYER ANGLE PREDICTED BY SEPARATION CONDITIONS IS ,F9.4,4H DEG/17X,84HTHE MAXIMUM UPPER SURFACE PRANDTL-MEYER ANGLE PREDICTED BY SEPARATION CONDITIONS IS ,F9.4,4H DEG)
WRITE (6,1000) SSMIN, SMIN, SMOUT, SSOUT

1000 FORMAT (/25X, 9HM(IN) = ,F8.4,2X,9HM(OUT) = 1,F8.4,5X,9HM(IN) = ,F8.4,10X,9HM(OUT) = ,F8.4)
WRITE (6,1001) RLOW, SSMLOW, SMLOW, SSLOW

1001 FORMAT (/25X, 9HM*(LOW) = ,F8.4,5X,9HM*(LOW) = ,F8.4,2X,9HM*(LOW) = ,F8.4,5X,10X,9HM*(LOW) = ,F8.4)
WRITE (6,1002) THETA, GSTAR, CSTAR, SIGMA

1002 FORMAT (/11X,84HTHETA = ,F8.4,4H DEG,12X,5HC* = ,F8.4,11X,8HSIGMA = ,F8.4)

IF (NPLOI .NE. 0) GO TO 1
CC IF PLOTMY IS NOT AVAILABLE, REMOVE THE FOLLOWING CARDS
C****MULTIPLE PLOT - START
LLL = NP1 + NP2
CALL SORTXY (XDOWN(1), YACRS1(1), NP1)
CALL SORTXY (XDOWN(NP1+1), YACRS1(NP1+1), NP2)
CALL SORTXY (XDOWN(LLL+1), YACRS1(LLL+1), NP3)
P(1) = 5.0
P(3) = 12.0
P(4) = 20.0
P(11) = ((I.*AMINI(YACRS1(1), YACRS1(NP1+1), YACRS1(LLL+1))) / 100.)*I(10,**4)
P(6) = 2.0
P(7) = AMINI(XDOWN(1), XDOWN(NP1+1), XDOWN(LLL+1)) *(I(10,**4)
P(8) = P(11)*5./3.)
P(9) = 2.0
P(10) = AMINI(YACRS1(1), YACRS1(NP1+1), YACRS1(LLL+1)) *(I(10,**4)
KKK(1) = 55
KKK(2) = 4
KKK(3) = NP1
KKK(5) = NP2
KKK(7) = NP3
KKK(9) = 1
DATA KKK(4), KKK(6), KKK(8), KKK(10) /1H#, 1H#, 1H#, 1H#, KKK(10)/1H/
CALL PLOTMY (XDOWN, YACRS1, KKK, P)
C****MULTIPLE PLOT - STOP
GO TO 1
END
SUBROUTINE ROOT (X0, X2, XINTL, FOFX, FUNC, XI)

COMMON/ROOTS/DELF
DOUBLE PRECISION X, XXO, XX2

C

C WE ARE SEEKING AN X SUCH THAT FUNC(X) = FOFX WHERE FOFX IS A KNOWN
C FUNCTIONAL VALUE
C
C 1 LOCATE FOFX IN (FO, FX) OR (FX, F2) WHERE FX IS THE PREVIOUS
C APPROXIMATION TO FOFX
C 2 LET X = 1/2(XX0 + X) OR X = 1/2(X + XX2)
C 3 IS FUNC(X) = FOFX = IF NOT, REPEAT PROCEDURE

XX0 = X0
XX2 = X2
FO = FUNC(XX0)
F2 = FUNC(XX2)

IF (FOFX .LT. FO .AND. FOFX .LT. F2 .OR. FOFX .GT. F0 .AND. FOFX .GT. F2)
GO TO 1005

IF (ABS(FOFX - F0) .LE. DELF) GO TO 1007
IF (ABS(FOFX - F2) .LE. DELF) GO TO 1008

X = XINTL
KOUNT = 0

1000 X1 = X
KOUNT = KOUNT + 1
A = FOFX - F2
FX = FUNC(X)

IF (KOUNT .GE. 60) WRITE (6, 1004) KOUNT, X, FX, FOFX

IF (ABS(FX - FOFX) .LE. DELF) RETURN
IF (KOUNT .EQ. 75) GO TO 1002
IF (A*(FX - FOFX) .LT. 0.) GO TO 1001

XX0 = X
X = (X + XX2)/2.
GO TO 1000

1001 XX2 = X
X = (XX0 + X)/2.
F2 = FX
GO TO 1000

1002 WRITE (6, 1003)
1003 FORMAT (/30X, 62H ITERATIONS HAVE BEEN PERFORMED WITHOUT CONVERG
LING TO A ROOT)
RETURN

1005 WRITE (6, 1006) FOFX
1006 FORMAT (/10X, 7HF(X) = , 6G16.9, 31H IS OUTSIDE OF SPECIFIED LIMITS)
RETURN

1007 XI = X0
RETURN

1008 XI = X2
RETURN

END
SUBROUTINE START (VLOW,FLO,VUP,FUP,VIMAX)

COMMON/FACTOR/PERM,SAME,GAM,GAMMI,GAMPI,SSMLOW,SSMUP,RECONV,BINTGR
EXTERNAL CFAC_,QFACT,FRAT,FOFRS,FKMAX

XO = 1./PERM
X2 = 0.9999999999
XINTL = (XO + X2)/2.

IF (VLOW .LT. 0.0) GO TO 70
CALL ROOT (XO,X2,XINTL,FLD,FOFRS,RLOW)
GO TO 71

70 RLOW = 1.0
71 SSMLOW = 1./RLOW
CALL ROOT (XO,X2,XINTL,FUP,FOFRS,RUP)
SSMUP = 1./RUP

IF (SSMLOW .LE. SSMUP) GO TO 40

C FKMAX(X) IS LINEAR IN A NEIGHBORHOOD OF X WHEN X IS SUCH THAT FKMAX(X)=0
C USE GOOD INITIAL ESTIMATE PLUS LINEARITY TO FIND X SUCH THAT FKMAX(X)=0

XINTL = (1./PERM)*SORT( SSMLOW/SSMUP )
X0 = XINTL - 0.005
F0 = FKMAX(X0,0)
X2 = XINTL + 0.001
F2 = FKMAX(X2,0)
SLOPE = (F2 - F0)/(X2 - XO)
FINTL = FKMAX(XINTL,0)
YINCPT = FINTL - SLOPE*XINTL
XAMK = -YINCPT/SLOPE
FOFX = FKMAX(XAMK,
IF (ABS(FOFX) .GT. 0.00009) WRITE (6,60) FOFX,XAMK

60 FORMAT (//29X,35HSEARCH FOR ROOT FAILED F(X) : ,G16,9,TH X = ,G16.9)
SAME = (XAMK/SSMLOW)*(XAMK/SSMLow)
C = 1. - PERM*(GAMPI**(1./GAM-1.))*(SSMUP/(SSMUP-SSMLOW))*XAMK*
BINTGR
CINTGR = SIMPS1(SSMLOW,SSMUP,QFACT,K)
Q = (SSMLOW*SSMUP/(SSMUP-SSMLOW))*CINTGR
RATIO = Q/(1. - C)
GO TO 50

40 XAMK = 1./PERM
RATIO = SSMUP*SSMUP*QFACT(SSMUP)
C = 0.0
Q = 0.0
GO TO 50

50 XO = 1.0
X2 = PERM
XINTL = (XO + X2)/2.
CALL ROOT (XO,X2,XINTL,RATIO,FRAT,SSMIA)
VIMAX = (3.14159265/4.)*(PERM-1.) + (PERM/2.)*AR Sin(2.*GAMMI*
1. SSMIA*SSMIA - GAM) + 0.5*AR Sin(2.*GAMPI/(SSMIA*SSMIA) - GAM)
\begin{verbatim}
VIMAX = VIMAX*RECONV
VLOW = VLOW*RECONV
VUP = VUP*RECONV
WSTAR = ((1./GAMPI)**(GAMPI/(2.*GAMMI)))*CINTGR

WRITE (6,10)
10 FORMAT (/48X,36Hcalculations for supersonic starting)

WRITE (6,90) WSTAR
90 FORMAT (/50X,24Hweight-flow parameter = ,F9.4)

WRITE (6,20) XAMK,C,O,SSMIX
20 FORMAT (/2OX,10H*(MAX) = ,F9.4,SX,5HC = ,F9.4,SX,5HO = ,F9.4,SX,113HM*(I(MAX)) = ,F9.4)

WRITE (6,30) VIMAX,VLOW,VUP,GAM
30 FORMAT (/4X,38Hthe maximum design value for v(in) is ,F9.4,21H deg
1. when v(low) is ,F9.4,16H deg, v(up) is ,F9.4,16H deg, gamma is
2. ,F7.4)
RETURN
END

$IBFTC MSS LIST

SUBROUTINE MSSTAR (M,N,VSSIAR)

COMMON/FACTOR/PERM,SAME,GAM,GAMMI,GAMPI,SSMLOW,SSMUP,RECONV,D
REAL M,MS
EXTERNAL ADSTR

A = 0.785398162*(PERM -1.)
B = 0.5*PERM
C = GAM -1.
D = GAM + 1.
IF(N .NE. 0) GO TO 1
MS = ADSTR(M)
1 GO TO 2
X0 = 1.
X2 = PERM
XINTL = (X0 + X2)/2.
FOFX = M
CALL ROOT (X0,X2,XINTL,FOFX,ADSTR,MS)
SQRDM = MS*MS
VSTSTAR = ( A + B*AR SIN(C*SQRDM-GAM) + 0.5*AR SIN(D/SQRDM-GAM) )*RECONV
RETURN

3 VSTSTAR = 0.
RETURN
END
\end{verbatim}
FUNCTION SIMPS1(XMIN, XMAX, FUNCI, KER)
DIMENSION V(200), H(200), A(200), B(200), C(200), P(200), E(200), NE(200)
EQUIVALENCE (E, NE), (TEST, NTEST)
T=3.0E-5
V(1)=XMIN
H(1)=0.5*(XMAX-XMIN)
A(1)=FUNCI(XMIN)
B(1)=FUNCI(XMIN+H(1))
C(1)=FUNCI(XMAX)
P(1)=H(1)*(A(1)+4.0*B(1)+C(1))
E(1)=P(1)
ANS=P(1)
N=1
FRAC=2.0*T
1 FRAC=0.5*FRAC
2 TEST=ABS(FRAC*ANS)
3 DO 7 I=1, K
4 IF (TEST-1ABS(NE(I))) 5,5,7
5 N = N+1
6 IF (N-200) 7, 13, 13
7 CONTINUE
8 IF (N-K) 9,9,2
9 Q = 0.0
10 DO 11 I=1, N
11 Q=Q+E(I)
12 IF (ABS(Q)-T*ABS(ANS)) 14,14,1
13 KER=KER+1
14 ANS=0.0
15 DO 16 I=1, N
16 ANS=ANS+P(I)
SIMPS1=(ANS+Q/3.0)/3.0
17 RETURN
END
$IBFTC ALLFUN
FUNCTION ALFUNC (A,B,Y)

COMMON/EXPALF/GAMEXP
COMMON/FACTOR/PERM,SAME,GAM,GAMMI,GAMPI,SSMLOW,SSMUP,RECONV,GRTY

ALFUNC = (1./Y)*(((A - B*Y*Y)**GAMEXP))

RETURN
END

$IBFTC BAKE
FUNCTION CFACT (Y)

COMMON/FACTOR/PERM,SAME,GAM,GAMMI,GAMPI,SSMLOW,SSMUP,RECONV,GRTY
EXTERNAL ALFUNC

CFACT = ALFUNC(1.,SAME,Y)

RETURN
END

$IBFTC CHARL
FUNCTION OFACT(Y)

COMMON/FACTOR/PERM,SAME,GAM,GAMMI,GAMPI,SSMLOW,SSMUP,RECONV,GRTY
EXTERNAL ALFUNC

OFACT = ALFUNC(GAMPI,GAMMI,Y)

RETURN
END

$IBFTC DOGG
FUNCTION FRAT(Y)

COMMON/FACTOR/PERM,SAME,GAM,GAMMI,GAMPI,SSMLOW,SSMUP,RECONV,GRTY
EXTERNAL OFACT,ALFUNC

FRAT = (Y**(GAM/GAMMI))*OFACT(Y)/ALFUNC(-GAMMI,-GAMPI,Y)

RETURN
END
FUNCTION FOFRS (X)

COMMON/FACTOR/PERM,SAME,GAM,GAMM1,GAMPI,SSMLOW,SSMUP,RECONV,GRTY
DOUBLE PRECISION X

ARG1 = 2.*GAMM1/(X*X) - GAM
ARG2 = 2.*GAMPI*X*X - GAM
IF (ABS(ARG1) .GT. 1.0 .OR. ABS(ARG2) .GT. 1.0) WRITE (6,1) ARG1,ARG2
1 FORMAT (//14X,61HARGUMENT OF ARCSIN IS OUTSIDE DOMAIN OF DEFINITION
IN ARG1 = ,G16.9; ARG2 = ,G16.9)

FOFRS = PERM*ARSGN(ARG1) + ARSIN(ARG2)

RETURN
END

FUNCTION FKMAX(Y,L)

COMMON/FACTOR/PERM,SAME,GAM,GAMM1,GAMPI,SSMLOW,SSMUP,RECONV,BINTGR
EXTERNAL ALFUNC,CFACT

SAME = (Y/SSMLOW)*(Y/SSMLOW)
K = 0
FKMAX = SIMPSI(SSMLOW,SSMUP,CFACT,K)
IF (K .EQ. 1) WRITE (6,1)
1 FORMAT (/10X,26HFAILURE TO INTEGRATE CFACT)
IF (L .EQ. 1) BINTGR=FKMAX

FKMAX = FKMAX + SSMUP*CFACT(SSMUP) - ALFUNC(Y,Y,Y,1.)

RETURN
END
FUNCTION ASTR(MSTAR)

COMM/N/FACTOR/PERM,SAME,GAM,GAM1,GAMPI,SSMLM,SSMUP,RECONV,D
REAL MSTAR,M

C = GAM - 1.
F = C/1)
G = C/GAM
H = GAM/1
M = MSTAR=MSTAR

ASTR = PERM*SORT( (1.-(1.-F*M))*((1.+0.5*((H*M)/(1.-F*M)))*G) )

RETURN
END

Lewis Research Center,
National Aeronautics and Space Administration,
Cleveland, Ohio, October 6, 1967,
128-31-02-25-22.

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