NASA TECHNICAL NOTE

NASA TN D-5805

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NASA TN D-5805

BUFFET AND STATIC AERODYNAMIC CHARACTERISTICS OF A SYSTEMATIC SERIES OF WINGS DETERMINED FROM A SUBSONIC WIND-TUNNEL STUDY

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NATIONAL AERONAUTICS AND SPACE ADMINISTRATION • WASHINGTON, D. C. • JUNE 1970

1. Report No.	2. Government Accession No.	3. Recipient's Catalog No.
NASA TN D-5805		

4. Title and Subtitle
BUFFET AND STATIC AERODYNAMIC CHARACTERISTICS OF A SYSTEMATIC SERIES OF WINGS DETERMINED FROM A SUBSONIC WIND-TUNNEL STUDY

5. Report Date
June 1970

6. Performing Organization Code

7. Author(s)
Edward J. Ray and Robert T. Taylor

8. Performing Organization Report No.
L-7011

9. Performing Organization Name and Address
NASA Langley Research Center
Hampton, Va. 23365

10. Work Unit No.
126-14-12-02-23

11. Contract or Grant No.

12. Sponsoring Agency Name and Address
National Aeronautics and Space Administration
Washington, D.C. 20546

13. Type of Report and Period Covered
Technical Note

14. Sponsoring Agency Code

15. Supplementary Notes

16. Abstract

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17. Key Words (Suggested by Author(s))
Buffet
Static aerodynamic characteristics
Camber
Aspect ratio
Sweep
Thickness-to-chord ratio
Position of maximum thickness

18. Distribution Statement
Unclassified – Unlimited

19. Security Classif. (of this report)	20. Security Classif. (of this page)	21. No. of Pages	22. Price*
Unclassified	Unclassified	265	$3.00

*For sale by the Clearinghouse for Federal Scientific and Technical Information
Springfield, Virginia 22151

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SUMMARY

A wind-tunnel investigation has been conducted in the Langley high-speed 7- by 10-foot tunnel to determine the buffet and static aerodynamic characteristics of a systematic wing series at Mach numbers ranging from 0.23 to 0.94. The results have indicated that for a given Mach number the wings which display superior aerodynamic efficiency characteristics generally display the highest buffet-free lift coefficient. The characteristics exhibited by the wings which were considered have indicated that correlations can be made between the onset of buffeting and selected divergences in the static aerodynamic characteristics. Axial force has been found to be the most sensitive static component to the onset of buffeting.

INTRODUCTION

The maneuverability and performance of aircraft engaged in air-to-air combat at high subsonic speeds are limited by the flow separation on the wing which manifests itself in a buffeting of the airframe and pronounced increases in drag. There are several approaches which the designer of new aircraft may employ in order to alleviate buffeting and its effects. An obvious method is the use of low wing loadings; however, this approach is limited by such considerations as cruise performance, structural weight, gust response, and so forth. A more desirable approach would be to determine methods of increasing the lift coefficient at which buffeting occurs by proper selection of planform, airfoil section, and variable-geometry devices.

As a contribution to the information needed for a proper selection of wing design parameters such as planform and airfoil section, a research program has been conducted to study the effects of systematic variations in wing design parameters on buffeting tendencies. The primary method for determining buffeting onset in this study has been by the wing-root bending-gage technique. However, another objective of this study was to evaluate other methods of determining buffet onset, such as particular variations in the static aerodynamic characteristics.

The study made use of 11 buffet models covering systematic variations of sweep, thickness-to-chord ratio, position of maximum thickness, camber, and aspect ratio. The study has been conducted over a range of Mach number from a minimum of about 0.23 to a maximum of 0.94. The purpose of this paper is to present the results of this buffet research program and to interpret the various wind-tunnel measurements with respect to buffeting.

SYMBOLS

The coefficients of forces and moments for the plotted longitudinal aerodynamic results are referred to the stability axis system with the exception of the axial-force and normal-force coefficients, which are referred to the body axis system. In addition to the plotted presentation, tabulations of the static longitudinal and lateral characteristics utilizing both body and stability axis systems are presented herein. The static aerodynamic forces and moments have been nondimensionalized by using the individual geometric characteristics of each wing (shown in table I). Pitching moments are referred to the quarter-chord point of the individual wing mean geometric chords.

The units used for the physical quantities in this report are given both in the U.S. Customary Units and in the International System of Units (SI). Factors relating the two systems are given in reference 1.

A	aspect ratio
b	wing span, in. (cm)
c	local chord of wing, in. (cm)
$\bar{c}$	mean geometric chord, in. (cm)
c_r	wing root chord, in. (cm)
c_t	wing tip chord, in. (cm)
C_A	axial-force coefficient, $\frac{\text{Axial force}}{qS}$
$(C_A)_{\alpha=0}$	axial-force coefficient at angle of attack of 0^o

C_D drag coefficient, $\frac{\text{Drag}}{qS}$

$C_{D,i}$ theoretical induced-drag coefficient

C_L lift coefficient, $\frac{\text{Lift}}{qS}$

C_{L_α} lift-curve slope (near angle of attack of 0^o)

$C_{L_{(L/D)_{max}}}$ lift coefficient at maximum lift-drag ratio

C_m pitching-moment coefficient, $\frac{\text{Pitching moment}}{qS\bar{c}}$

$C_{m_{C_L}}$ static margin (taken at low-lift coefficients), $\frac{\partial C_m}{\partial C_L}$

C_N normal-force coefficient, $\frac{\text{Normal force}}{qS}$

L/D lift-drag ratio

M_{WSG} root-mean-square moment of wing bending gage, in. lb (m-N)

q free-stream dynamic pressure, lb/ft^2 (N/m^2)

r radius, in. (cm)

R Reynolds number per foot (per meter)

S wing reference area, ft^2 (m^2)

s ratio of actual to theoretical leading-edge suction force (ref. 2)

α angle of attack, deg

$\Lambda_{c/4}$ wing quarter-chord sweepback, deg

Subscript:

max	maximum

ABBREVIATIONS

FS	fuselage station (measured from nose of model), in. (cm)
L.E.	leading edge
rms	root mean square

MODELS

A two-view sketch of the general model arrangement is shown in figure 1. The reference center of gravity was assumed to be in the plane of symmetry of the fuselage at the quarter-chord point of the wing mean geometric chords. The geometric characteristics of the various wings which were studied in this investigation are presented in table I.

Eleven buffet wings were used in the study which covered systematic variations of sweep, thickness-to-chord ratio, position of maximum thickness, camber, and aspect ratio. Each wing was constructed of a solid SAE 4130 steel panel, and particular care was taken to insure that the steel wings were rigidly attached to the steel portion of the fuselage to minimize structural damping. The buffet gages, which constituted a complete moment bridge of four active strain gages, were embedded beneath the upper and lower surface of one wing panel near the fuselage juncture on the 50-percent-chord line. (See fig. 1.) The recesses over the gages were filled in and faired to the contour of the wing surface. All wiring was routed internally through the model into the balance chamber.

Most of the wings were tested in combination with the fuselage having the rounded forebody. Several tests, however, were performed with a fuselage having a pointed forebody (see fig. 1) to determine whether the buffet-onset or buffet-intensity characteristics were affected by forebody bluntness.

MEASUREMENTS AND CORRECTIONS

Measurement of Buffet Characteristics

The primary source of buffet information for this investigation was obtained by the wing-root bending-gage technique (ref. 3). The buffet gages consisted of four active

strain gages forming a complete bending-moment bridge. As shown in figure 1, the gages were located near the wing-fuselage juncture and oriented along the 50-percent-chord line of the wing. The gages located in this manner, that is, on or near the effective flexural and cantilever axes, respond readily to any fluctuating aerodynamic load disturbance on the wing panel due to flow separation.

Prior to the wind-on investigation of the buffet models, each wing was oscillated with a vibrator through a range of frequencies varying from about 0 to 600 Hz to determine the bending characteristics of the wings and to insure that the gages were sensitive to oscillations at the fundamental bending frequency. The equipment used in this static survey included an oscillator with associated electronic equipment, a hand-held vibration pickup, and an oscilloscope. The models were driven through the known frequency range, and the phase relationship and amplitude of the oscillator and vibrator pickup were monitored on the oscilloscope. Because the vibration probe could be held at various locations on the models, the fundamental bending frequencies could be identified, and the nodal patterns of the various buffet wings could be determined.

At the outset of this investigation, the onset of buffeting was determined by visually monitoring the wind-on root-mean-square outputs of the wing-bending gages on a simple root-mean-square voltmeter (test 768). This method for determining buffet onset appeared to be adequate in most cases; however, at the higher values of gage output it became difficult to ascertain a quantitative level of buffet intensity because of large fluctuations of the root-mean-square meter. This was particularly true in the high-subsonic Mach number range where the rise in buffet intensity appeared to become a more gradual process in contrast to the well-defined, abrupt onset of buffeting indicated at the lower Mach numbers.

In order to establish a higher degree of repeatability of the buffet responses, a root-mean-square meter which linearly converts alternating-current input into direct-current outputs (test 778) and a direct-current integrator were incorporated into the system. The final arrangement of the buffet instrumentation is shown in figure 2.

A brief study was conducted to determine a realistic integration time with respect to available test time and repeatability of the buffet results. The bending-gage outputs were monitored on strip charts for intervals ranging from 2 minutes to 30 seconds at several angles of attack and Mach numbers. It was determined that for a typical test wing at several different tunnel conditions, an acceptable degree of repeatability could be obtained by integrating the output of the bending gages for a period of about 45 seconds. Therefore, the buffet results presented herein (with the exception of the results for wing 4) represent the average root-mean-square values of the alternating currents emitted from the wing moment bridge during a 45-second sampling interval. (The buffet

results shown for wing 4 (test 768, see fig. 7(g)) were determined by the earlier method which relied on visual observation of the simple-root-mean-square meter.)

Measurements of Static Aerodynamic Forces and Moments

The static aerodynamic forces and moments were measured by means of a six-component electrical strain-gage balance which was installed within the model. The static aerodynamic data were recorded simultaneously with the integrated root-mean-square output of the wing bending gages.

Transition strips of No. 150 carborundum grit were placed 0.50 inch (1.27 cm) behind the leading edges of the wings and 1.00 inch (2.54 cm) aft of the fuselage nose in the manner described in reference 4 to insure turbulent flow in the model boundary layer at Mach numbers above approximately 0.50. It should be emphasized here that several studies were made with transition at Mach numbers below 0.50, and therefore, the drag results at the low subsonic Mach numbers should be used with caution. Wings 8 and 9 were investigated with the transition strips completely removed to determine the effects of the artificial roughness on the buffet and static aerodynamic characteristics.

Corrections to Static Aerodynamic Results

The angles of attack shown herein have been corrected for the combined bending of the sting and balance system due to aerodynamic loading. Balance cavity pressures were monitored throughout the investigation by means of differential pressure gages and the axial-force and drag-coefficient data have been adjusted to correspond to a condition of free-stream static pressure at the base of the model. Jet boundary and blockage corrections were applied to the results as prescribed in references 5 and 6.

TEST CONDITIONS

The investigation was conducted in the Langley high-speed 7- by 10-foot tunnel which is a continuous-flow facility having, for this study, a closed test section. The Mach numbers for the various tests performed are listed in the run schedule contained in table II. The variations of the average Reynolds number and dynamic pressure with Mach number are shown in figure 3. In general, the Mach number range of this investigation extended from a minimum of about 0.23 to a maximum of about 0.94. The models were tested at 0° of sideslip through an angle-of-attack range which was varied from about 0 to a maximum of about 22°. At the higher Mach numbers, the angle-of-attack range was reduced because of balance limitations.

RESULTS AND DISCUSSION

Presentation of Results

Table II presents a run schedule for the buffet and static aerodynamic results presented in table III. Results are tabulated for all the wings which have been investigated with the exception of the buffet results for wing 4. (The buffet data for wing 4, determined in a preliminary investigation, were recorded and computed manually and therefore are not included in table III.)

Plotted results have been included for wings 1 to 11. (The wings were tested with transition grit on and with the fuselage having the rounded forebody except where specifically noted.) The results are presented in the following figures:

Evaluation of Buffet-Onset Measurement Techniques

Self-induced wing buffet is associated with the wing response to random excitation resulting from pressure fluctuations in separated flow from the wing. Separated-flow conditions on wind-tunnel models can be determined by a variety of techniques which include static force and moment measurements, such as inflections in the lift-curve slope; divergences in trailing-edge pressure; observations of the flow in the boundary layer; fluctuating wake pressure studies; and wing-root bending-gage measurements. All these methods have been utilized in the past with varying degrees of success to determine the conditions at which buffet onset occurs.

Since buffeting is a dynamic phenomenon, the fluctuating-root bending-gage technique appeared at the outset to be one of the most direct and simple methods for determining buffet onset in that it readily provides an integration of the oscillations occurring on the entire wing panel. Past research (ref. 7) has indicated that the wing-root bending-gage technique can be a reasonably effective method for predicting flight buffet-onset boundaries from wind-tunnel results. Insofar as the development of a particular configuration for good high-lift characteristics is concerned, the information provided by the wing-root bending gage is certainly recognized as being inadequate to determine where separation occurs and needs to be supplemented by other information, such as extensive pressure-distribution information and possibly, boundary-layer flow observations. There are further shortcomings in regard to the interpretation of the bending-gage results under certain conditions where the buffet-onset lift coefficient approaches zero. Usually this phenomenon occurs at conditions where the Mach number is high and the thickness-induced flow field is substantial (for instance, a wing with low sweep and high thickness-to-chord ratio). Under these conditions, self-induced buffeting may occur where separation exists on the wing at all lift coefficients. At the higher Mach numbers, which obviously are of particular interest in this type of study, this situation is complicated by a rise in the "noise," or wind-tunnel turbulence level. (Tunnel "noise" is manifested in the bending-gage results as an increase in output of the gage with Mach number which is insensitive to changes in model attitude.) Under conditions such as this, that is, low lift coefficient for buffet onset and substantially high levels of tunnel turbulence (for example, see the case for a Mach number of 0.883 in fig. 4(g)), the uncertainty exists as to whether

the wing-bending-gage output at the low lift coefficient is due to separation on the wing or due to the wind-tunnel turbulence. In cases such as this, where a well-defined "plateau," or base level, in the bending-gage results and a pronounced divergence are not established as the model attitude is increased, additional aid is required to interpret the results with any degree of certainty. In the present study, the buffet-onset characteristics of these more "complicated" cases were interpreted with the aid of the static-force data. Classically, erratic variations in the axial-force coefficient with angle of attack suggest separation on the wing of a wind-tunnel model. "Erratic" variations in this discussion are assumed to be departures of the experimental results from the theoretical variations of axial-force coefficient with angle of attack. (An indication of separation could also be obtained by evaluating the experimental axial-force variations with the square of the angle of attack. A linear variation of axial-force coefficient with the square of the angle of attack would suggest that the airflow is attached to the wing, and departures from this linear variation would indicate that the flow over the wing has separated.)

The basic results determined for the wing 1 configuration will be discussed to illustrate the process used in the determination of the buffet characteristics presented in this paper. In figure 4, the static longitudinal aerodynamic characteristics and wing-bending-gage moment M_{WSg} exhibited by the wing 1 configuration are presented as functions of angle of attack α or lift coefficient C_L. The solid symbols shown on each of the presentations represent the buffet-onset points which were determined, primarily, from the wing-bending-gage results. In reviewing the bending-gage results (fig. 4(g)), it will be noted that at the lower Mach numbers the rms signal of the wing bending-gage output M_{WSg} is essentially invariant with C_L until some well-defined break takes place. At the lower Mach numbers the condition at which the divergence occurs is obvious, and the determination of the buffet-onset points requires very little interpretation. As the Mach number is increased, the zero-lift level of the rms moment output is increasing before the buffeting occurs as a result of the increase in the tunnel turbulence level. At the highest Mach numbers, the results indicate a gradual rise in the bending-gage intensity prior to the abrupt rise. For example, in the Mach 0.88 case, the bending-gage output appears to have increased between a lift coefficient of about 0 to 0.15; however, there is a definite "break" in the curve at a lift coefficient of about 0.45. In this region of uncertainty $C_L = 0.15$ to 0.45, the wing possibly experiences a low-intensity, or intermittent, buffeting. Interpretation of this particular case is difficult and the exact lift coefficient at which buffet onset occurs cannot be definitely established from this information alone. For this case, a better indication of the point of buffet onset can be obtained from the axial-force results presented in figure 4(b). The theoretical axial-force curves (indicated by the dashed lines) were developed from the equation

$$C_A = s\left[C_{L_\alpha} - \left(C_{L_\alpha}\right)^2 \frac{\partial C_{D,i}}{\partial C_L^2}\right]\sin^2\alpha + \left(C_A\right)_{\alpha=0}$$

which was derived from information contained in references 2 and 8 and the experimental axial-force coefficients at zero angle of attack $\left(C_A\right)_{\alpha=0}$. The C_{L_α} and $\partial C_{D,i}/\partial C_L^2$ terms were calculated by a modified Multhopp lifting-surface theory developed in reference 9.

It will be noted from the axial-force results (fig. 4(b)) for Mach numbers of 0.82 and below, that the point at which the experimental data depart from the theoretical variations is in good agreement with the buffet-onset points, which are obvious from the bending-gage results (fig. 4(g)). These data suggest the correspondence between the deviation in the axial-force variations and the onset of buffeting. A rational explanation exists if separation in the boundary layer is presumed to cause each of these phenomenon. Acceptance of this argument leads to a very convenient method of complementing the wing-bending-gage results in analyzing buffet-onset characteristics.

Again, comparison of both the axial-force results (fig. 4(b)) and the bending-gage data (fig. 4(g)) shows that in the lower Mach number range there is a pronounced reversal of the experimental axial-force slopes which correlates with an abrupt divergence in the wing bending-gage output. At a Mach number of 0.88, where the interpretation of the bending-gage results becomes more difficult, the experimental axial-force results indicate a noticeable departure from the theoretical curves at an angle of attack of about 1^o which corresponds to the slight variation in the wing-bending-gage results at a lift coefficient of about 0.05. These results lead to the conclusion that separation did exist at the lower attitudes and that the gradual rise evidenced in the bending-gage output resulted from a low-intensity excitation (buffeting) of the wing panel. It should be emphasized here that this "conservative" approach has been taken in the analysis of all of the buffet-onset characteristics shown herein. (Although not included in the data presentations, the preliminary investigation previously mentioned indicated that at a Mach number of about 0.94 the wing 1 configuration exhibited buffeting at all lift coefficients.) The pronounced increase in buffet intensity noted for the Mach 0.88 condition at a lift coefficient of 0.45 ($\alpha = 4^o$) corresponds to the attitude at which the slope reverses in the experimental axial-force variation. The significance of this apparent onset of low-intensity buffeting is not fully understood at this point, but it could be conjectured that in the high-subsonic—transonic region where a transition from the classic subsonic separation to a shock-induced separation takes place, two different types of buffeting phenomena might occur at one Mach number. The low-intensity buffeting might possibly be indicative of the development of a shock-induced separation, whereas the pronounced increase in buffet

intensity could be associated with the type of subsonic separation and buffeting exhibited at the lower subsonic Mach numbers. Additional studies, possibly visual flow observations, might be of assistance in the understanding of these suspected low-intensity and high-intensity "boundaries."

Interpretation of the bending-gage results obtained at the higher subsonic Mach numbers is not always as difficult as the Mach 0.88 condition for wing 1 just discussed. As an illustration, the rms output of the wing gages in figure 8(g) for a Mach number of about 0.93 can readily be interpreted. The pronounced rise in the bending-gage output at low attitudes suggests that there is well-defined separation on the relatively thick wing 5 configuration and that the airfoil buffets at a very low lift coefficient, probably at a lift coefficient of zero.

Although the axial force appears to be a useful aid in determination of buffet-onset characteristics in selected cases of uncertainty, this paper does not intend to suggest that the wing bending gage is not useful, or required, in the analysis of buffet-onset characteristics. Erratic axial-force characteristics (i.e., departures of the experimental axial force from the theoretical trend) may be caused by the formation of a shock system which is of insufficient strength to separate the boundary layer; therefore, care must be taken when interpreting the axial-force variations. Self-induced buffeting, however, is not possible without separation.

All the static-force and moment characteristics were examined in conjunction with the bending-gage results to determine whether any additional correlations could be established. In general, for the uncambered series of wings, the break in the lift-curve slope, which is commonly used to predict buffet onset, appeared to indicate optimistic predictions, particularly in the low-subsonic Mach number range. However, for the cambered wings, which develop considerably higher buffet-free lift coefficients, a fair correlation appeared to exist between the breaks in the lift-curve and pitching-moment slopes and the wing-bending-gage divergences. (For example, see fig. 15.) The possibility of using the divergences in the rolling-moment curves as an indication of buffet onset was examined and the trends indicated were found to be inconsistent; therefore, a satisfactory correlation with the wing-bending-gage results could not be established. This approach was not expected to offer a dependable indication of buffet onset since the rolling-moment divergences at zero angle of sideslip would be dependent upon asymmetric separation characteristics.

Wing-Geometry Effects on Buffet Onset

The effects of the various wing parameters which were evaluated are illustrated at comparable Mach numbers in the comparison plots contained in figures 18 to 24. Figure 23 shows that the application of transition grit did not significantly affect the lift

coefficient for buffet onset of the wing 8 configuration. Sizable changes are evident, however, in the buffet intensity and static longitudinal characteristics for several Mach numbers. The bending-gage results shown in figure 24 at Mach numbers above 0.75 indicate that the addition of transition grit had a significant effect on the buffet-onset characteristics of the more highly cambered airfoil, wing 9. It is not conclusive from this brief study, but the effect of transition grit on buffet-onset characteristics appears to be dependent to some extent on the amount of camber in the airfoil sections. In addition, the contents of figure 24 indicate that the forebody-shape variations considered in this study did not significantly affect the buffet-onset characteristics of the wings. However, at the higher Mach numbers the results suggest that the shape of the forebody did affect the level of buffet intensity.

Summaries of the buffet characteristics and several static longitudinal parameters for the wings which were considered in this investigation are presented in figure 25. Several relationships evident in the variations of these parameters with Mach number are interesting when it is recalled that buffeting is a phenomenon which occurs in separated flow. There is a strong resemblance between the comparative buffet-onset trends and the variations of the maximum lift-drag ratios. At a given Mach number, the wing developing the highest maximum L/D and lift coefficient for maximum L/D generally displays the highest lift coefficient for buffet onset. It will be noted that, in general, when there is a divergence in the $C_{m_{C_L}}$ trend, a reduction in C_{L_α}, a marked increase in $(C_A)_{\alpha=0}$, and large reductions in $(L/D)_{max}$, the buffeting occurs at very low lift coefficients. (For instance, see the results for airfoil section 63A010 in fig. 25(b).)

In summary, the buffet-onset characteristics determined in this investigation have indicated that for the wing variations which were considered, the lower sweep, high-aspect-ratio, moderately thick, cambered airfoils with rounded leading edges should display higher buffet-free lift coefficients at low subsonic Mach numbers. In the high-subsonic regime, the results suggest that within the range of variables studied, increase in sweepback and reductions in thickness ratio, aspect ratio, and camber should favorably affect buffet-onset characteristics.

CONCLUDING REMARKS

A wind-tunnel investigation has been conducted in the Langley high-speed 7- by 10-foot tunnel to determine the buffet and static aerodynamic characteristics of a systematic wing series at Mach numbers ranging from 0.23 to 0.94. The results have indicated that for a given Mach number, the wings which display superior aerodynamic efficiency characteristics generally display the highest buffet-free lift coefficient. The

characteristics exhibited by the wings which were considered have indicated that correlations can be made between the onset of buffeting and selected divergences in the static aerodynamic characteristics. Axial force has been found to be the most sensitive static component to the onset of buffeting.

Langley Research Center,
National Aeronautics and Space Administration,
Langley Station, Hampton, Va., February 16, 1970.

REFERENCES

1. Mechtly, E. A.: The International System of Units – Physical Constants and Conversion Factors. NASA SP-7012, 1964.

2. Henderson, William P.: Studies of Various Factors Affecting Drag Due to Lift at Subsonic Speeds. NASA TN D-3584, 1966.

3. Davis, Don D., Jr.; and Huston, Wilber B.: The Use of Wind Tunnels To Predict Flight Buffet Loads. NACA RM L57D25, 1957.

4. Braslow, Albert L.; Hicks, Raymond M.; and Harris, Roy V., Jr.: Use of Grit-Type Boundary-Layer-Transition Trips on Wind-Tunnel Models. NASA TN D-3579, 1966.

5. Gillis, Clarence L.; Polhamus, Edward C.; and Gray, Joseph L., Jr.: Charts for Determining Jet-Boundary Corrections for Complete Models in the 7- by 10-Foot Closed Rectangular Wind Tunnels. NACA WR L-123, 1945. (Formerly NACA ARR L5G31.)

6. Herriot, John G.: Blockage Corrections for Three-Dimensional-Flow Closed-Throat Wind Tunnels, With Consideration of the Effects of Compressibility. NACA Rep. 995, 1950. (Supersedes NACA RM A7B28.)

7. Mabey, D. G.: Comparison of Seven Wing Buffet Boundaries Measured in Wind Tunnels and in Flight. C.P. No. 840, Brit. A.R.C., 1966.

8. Polhamus, Edward C.: A Concept of the Vortex Lift of Sharp-Edge Delta Wings Based on a Leading-Edge-Suction Analogy. NASA TN D-3767, 1966.

9. Lamar, John E.: A Modified Multhopp Approach for Predicting Lifting Pressures and Camber Shape for Composite Planforms in Subsonic Flow. NASA TN D-4427, 1968.

TABLE I.- MODEL CHARACTERISTICS

[Fuselage dimensions: length, 48.80 in. (123.95 cm); maximum diameter, 4.90 in. (12.45 cm); base diameter, 3.08 in. (7.82 cm)]

Wing	Airfoil section	$\Lambda_{c/4}$, deg	c_t, in. (cm)	c_r, in. (cm)	$\bar{c}$, in. (cm)	b, in. (cm)	S, ft^2 (m^2)	A	FS from L.E. of c_r, in. (cm)
1	NACA 63A008	25	3.43 (8.71)	8.57 (21.8)	6.38 (16.2)	36.00 (91.4)	1.50 (0.139)	6	19.27 (48.95)
2	NACA 63A008	35	3.43 (8.71)	8.57 (21.8)	6.38 (16.2)	36.00 (91.4)	1.50 (0.139)	6	18.87 (47.93)
3	NACA 63A008	45	3.43 (8.71)	8.57 (21.8)	6.38 (16.2)	36.00 (91.4)	1.50 (0.139)	6	18.34 (46.58)
4	NACA 63A006	35	3.43 (8.71)	8.57 (21.8)	6.38 (16.2)	36.00 (91.4)	1.50 (0.139)	6	18.87 (47.93)
5	NACA 63A010	35	3.43 (8.71)	8.57 (21.8)	6.38 (16.2)	36.00 (91.4)	1.50 (0.139)	6	18.87 (47.93)
6	NACA 64A008	35	3.43 (8.71)	8.57 (21.8)	6.38 (16.2)	36.00 (91.4)	1.50 (0.139)	6	18.87 (47.93)
7	NACA 65A008	35	3.43 (8.71)	8.57 (21.8)	6.38 (16.2)	36.00 (91.4)	1.50 (0.139)	6	18.87 (47.93)
8	NACA 63A208	35	3.43 (8.71)	8.57 (21.8)	6.38 (16.2)	36.00 (91.4)	1.50 (0.139)	6	18.87 (47.93)
9	NACA 63A408	35	3.43 (8.71)	8.57 (21.8)	6.38 (16.2)	36.00 (91.4)	1.50 (0.139)	6	18.87 (47.93)
10	NACA 63A008	35	4.20 (10.7)	10.50 (26.7)	7.80 (19.8)	29.39 (74.7)	1.50 (0.139)	4	18.80 (47.75)
11	NACA 63A008	35	3.76 (9.55)	9.39 (23.9)	6.98 (17.7)	32.86 (83.5)	1.50 (0.139)	5	18.84 (47.85)

TABLE II.- RUN SCHEDULE

Run	Wing	Nose	Mach number	Transition grit
		Test 768		
23	4	Round	0.940	On
24	↓	↓	.870	↓
25			.811	
26			.758	
27			.707	
28			.503	
		Test 778		
1	9	Round	0.933	Off
2	↓	↓	.863	↓
3			.808	
4			.753	
5			.702	
6			.498	
7			.292	
8		Pointed	.933	
9		↓	.863	
10			.807	
11			.753	
12			.702	
13			.498	
14	↓		.292	↓
15	8	Round	.833	On
16	↓	↓	.863	↓
17			.807	
18			.754	
19			.702	
20			.498	
21			.292	
22			.807	Off
23			.753	↓
24			.702	
25			.498	
26			.293	
27	↓		.932	
28			.863	
29	7		.932	On
30	↓		.863	↓
31			.807	
32			.754	
33			.702	
34			.498	
35	↓		.292	
36	5		.933	
37	↓		.864	
38			.806	
39			.754	
40			.702	↓

TABLE II.- RUN SCHEDULE – Continued

Run	Wing	Nose	Mach number	Transition grit
Test 778 – Continued				
41	5	Round	0.493	On
42	5		.293	
43	6		.933	
44			.864	
45			.806	
46			.755	
47			.702	
48			.293	
49			.497	
50	10		.933	
51			.864	
52			.807	
53			.755	
54			.703	
55			.293	
56			.498	
57	11		.932	
58			.863	
59			.807	
60			.754	
61			.702	
62			.293	
--	--	-----	-----	--
64	11	Round	.497	On
65	9		.883	
66			.823	
67			.770	
68			.718	
69			.666	
70			.225	
71			.455	
72	1		.883	
73			.823	
74			.770	
75			.718	
76			.665	
77			.225	
78			.456	
79	3		.883	
80			.821	
81			.769	
82			.717	
83			.665	
84			.225	
85			.455	

TABLE II.- RUN SCHEDULE – Concluded

Run	Wing	Nose	Mach number	Transition grit
Test 778 – Concluded				
86	2	Round	0.883	On
87	↓	↓	.770	↓
88			.717	
89			.665	
90			.822	
91			.456	
92			.225	

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS

The symbols used in the tabulated data are defined as follows:

MACH	Mach number
Q	free-stream dynamic pressure, lb/ft^2 (1 lb/ft^2 = 47.88 N/m^2)
BETA	angle of sideslip, deg
ALPHA	angle of attack, deg

Body axis:

CNF	normal-force coefficient
CAF	axial-force coefficient
CLB	rolling-moment coefficient
CNB	yawing-moment coefficient
CSF	side-force coefficient
CAB + CAC	chamber axial-force coefficient

Stability axis:

CL	lift coefficient
CD	drag coefficient
CPM	pitching-moment coefficient
CLS	rolling-moment coefficient
CNS	yawing-moment coefficient
L/D	lift-to-drag ratio
CDB + CDC	chamber drag coefficient
PB-1	integrated rms output (M_{WSg} on plotted presentations) from wing bending gage, in. lb (1 in. lb = 0.113 m-N)

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 768 RUN= 23

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
538	0.936	737.243	-0.01	-1.07	-0.1591	0.0162	-0.0011	0.0007	0.0002	0.0031	0.0031
539	0.939	739.626	-0.01	0.12	-0.0263	0.0174	-0.0006	0.0006	0.0004	0.0031	0.0031
540	0.940	740.409	-0.00	1.40	0.1237	0.0167	-0.0000	0.0005	-0.0006	0.0031	0.0031
541	0.939	740.015	-0.01	2.58	0.2595	0.0143	-0.0003	0.0005	0.0004	0.0031	0.0031
542	0.939	739.685	-0.01	3.82	0.3906	0.0115	-0.0004	0.0005	0.0003	0.0031	0.0031
543	0.940	740.258	-0.01	5.00	0.5100	0.0099	-0.0009	0.0005	0.0006	0.0031	0.0030
544	0.940	740.261	-0.01	6.22	0.6173	0.0085	-0.0010	0.0005	0.0003	0.0030	0.0030
545	0.943	742.844	-0.01	7.41	0.7169	0.0087	0.0005	0.0004	0.0007	0.0030	0.0030
546	0.941	741.527	-0.01	8.56	0.7651	0.0076	0.0025	0.0005	0.0046	0.0030	0.0029
547	0.937	738.187	-0.01	0.08	-0.0244	0.0170	-0.0001	0.0006	-0.0001	0.0031	0.0031
553	0.943	742.632	-0.01	0.12	-0.0226	0.0176	-0.0008	0.0007	0.0005	0.0032	0.0031
554	0.938	738.403	-0.00	-1.10	-0.1561	0.0161	-0.0006	0.0006	-0.0012	0.0031	0.0031
555	0.937	737.594	-0.01	0.07	-0.0285	0.0174	-0.0004	0.0006	0.0002	0.0031	0.0031

TEST= 768 RUN= 23

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	CDC
538	0.936	737.243	-0.01	-1.07	-0.1588	0.0191	0.0139	-0.0011	0.0007	-8.306	0.0031	0.0031
539	0.939	739.626	-0.01	0.12	-0.0263	0.0174	0.0083	-0.0006	0.0006	-1.515	0.0031	0.0031
540	0.940	740.409	-0.00	1.40	0.1233	0.0197	0.0303	0.0000	0.0005	6.244	0.0031	0.0031
541	0.939	740.015	-0.01	2.58	0.2586	0.0260	-0.0143	-0.0003	0.0005	9.963	0.0031	0.0031
542	0.939	739.685	-0.01	3.82	0.3889	0.0374	-0.0312	-0.0003	0.0005	10.390	0.0031	0.0031
543	0.940	740.258	-0.01	5.00	0.5072	0.0543	-0.0454	-0.0008	0.0005	9.338	0.0030	0.0030
544	0.940	740.261	-0.01	6.22	0.6127	0.0753	-0.0480	-0.0010	0.0006	8.136	0.0030	0.0030
545	0.943	742.844	-0.01	7.41	0.7098	0.1011	-0.0567	0.0005	0.0004	7.019	0.0030	0.0030
546	0.941	741.527	-0.01	8.56	0.7752	0.1244	-0.0286	0.0026	0.0002	6.231	0.0029	0.0029
547	0.937	738.187	-0.01	0.08	-0.0244	0.0170	0.0074	-0.0001	0.0006	-1.436	0.0031	0.0031
553	0.943	742.632	-0.01	0.12	-0.0226	0.0175	0.0081	-0.0008	0.0007	-1.290	0.0032	0.0031
554	0.938	738.403	-0.00	-1.10	-0.1558	0.0191	0.0147	-0.0006	0.0006	-8.170	0.0031	0.0031
555	0.937	737.594	-0.01	0.07	-0.0286	0.0174	0.0087	-0.0004	0.0006	-1.641	0.0031	0.0031

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 768 RUN= 24

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
556	0.868	682.237	-0.00	-1.10	-0.1338	0.0143	-0.0006	0.0006	-0.0011	0.0027	0.0026
557	0.867	681.598	-0.01	0.04	-0.0256	0.0162	-0.0002	0.0006	0.0000	0.0027	0.0026
558	0.867	681.609	-0.00	1.22	0.0859	0.0152	-0.0002	0.0006	-0.0005	0.0027	0.0026
559	0.870	683.969	-0.00	2.42	0.2073	0.0108	-0.0003	0.0006	-0.0000	0.0027	0.0027
560	0.869	683.617	-0.01	3.67	0.3367	0.0040	-0.0003	0.0006	0.0004	0.0026	0.0027
561	0.868	681.985	-0.01	4.88	0.4735	-0.0024	0.0000	0.0006	0.0007	0.0026	0.0026
562	0.869	683.473	-0.01	6.13	0.6102	-0.0051	0.0008	0.0007	0.0007	0.0026	0.0026
563	0.869	683.367	0.00	7.29	0.6558	0.0013	-0.0057	-0.0003	0.0011	0.0026	0.0026
564	0.871	685.057	0.00	8.44	0.7129	0.0053	-0.0054	-0.0003	0.0012	0.0025	0.0025
565	0.870	683.813	-0.00	9.53	0.7367	0.0082	-0.0017	0.0001	0.0012	0.0025	0.0025
566	0.871	684.883	-0.00	10.58	0.7569	0.0119	-0.0004	0.0003	0.0005	0.0025	0.0025
567	0.872	685.419	-0.00	11.71	0.7872	0.0141	-0.0000	0.0003	0.0008	0.0025	0.0024
568	0.873	686.564	-0.00	12.78	0.8293	0.0156	-0.0005	0.0004	0.0008	0.0024	0.0024
569	0.873	686.544	-0.01	13.91	0.8850	0.0156	-0.0002	0.0004	0.0012	0.0024	0.0024
570	0.869	682.879	-0.00	16.17	0.9581	0.0160	0.0002	0.0004	0.0010	0.0022	0.0023
571	0.876	688.805	-0.00	18.28	1.0295	0.0151	0.0003	0.0005	0.0003	0.0021	0.0020
572	0.877	690.100	-0.00	19.35	1.0796	0.0146	0.0003	0.0005	0.0001	0.0019	0.0019

TEST= 768 RUN= 24

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	CDC
556	0.868	682.237	-0.00	-1.10	-0.1335	0.0168	0.0034	-0.0006	0.0006	-7.935	0.0027	0.0026
557	0.867	681.598	-0.01	0.04	-0.0256	0.0162	0.0069	-0.0002	0.0006	-1.579	0.0027	0.0025
558	0.867	681.609	-0.00	1.22	0.0855	0.0171	0.0096	-0.0002	0.0007	5.014	0.0027	0.0026
559	0.870	683.969	-0.00	2.42	0.2067	0.0196	0.0121	-0.0002	0.0006	10.563	0.0027	0.0027
560	0.869	683.617	-0.01	3.67	0.3357	0.0256	0.0138	-0.0003	0.0006	13.128	0.0026	0.0026
561	0.868	681.985	-0.01	4.88	0.4719	0.0379	0.0076	0.0001	0.0006	12.454	0.0026	0.0026
562	0.869	683.473	-0.01	6.13	0.6072	0.0601	-0.0014	0.0009	0.0006	10.104	0.0026	0.0026
563	0.869	683.367	0.00	7.29	0.6504	0.0845	0.0371	-0.0057	0.0004	7.696	0.0026	0.0025
564	0.871	685.057	0.00	8.44	0.7044	0.1098	0.0551	-0.0054	0.0005	6.416	0.0025	0.0025
565	0.870	683.813	-0.00	9.53	0.7252	0.1301	0.0808	-0.0017	0.0004	5.573	0.0025	0.0025
566	0.871	684.883	-0.00	10.58	0.7419	0.1506	0.0963	-0.0003	0.0004	4.926	0.0025	0.0024
567	0.872	685.419	-0.00	11.71	0.7680	0.1736	0.1070	0.0000	0.0003	4.425	0.0024	0.0024
568	0.873	686.564	-0.00	12.78	0.8053	0.1986	0.1152	-0.0004	0.0005	4.055	0.0024	0.0023
569	0.873	686.544	-0.01	13.91	0.8358	0.2232	0.1271	-0.0001	0.0004	3.745	0.0023	0.0023
570	0.869	682.879	-0.00	16.17	0.9158	0.2821	0.1437	0.0003	0.0003	3.246	0.0021	0.0022
571	0.876	688.805	-0.00	18.28	0.9729	0.3372	0.1463	0.0005	0.0004	2.885	0.0020	0.0019
572	0.877	690.100	-0.00	19.35	1.0138	0.3715	0.1469	0.0004	0.0003	2.729	0.0018	0.0018

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 768 RUN= 25

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
573	0.812	632.386	-0.01	-1.07	-0.1249	0.0139	-0.0007	0.0006	0.0011	0.0024	0.0024
574	0.810	630.827	-0.01	0.05	-0.0198	0.0160	-0.0005	0.0006	0.0003	0.0024	0.0024
575	0.811	631.332	-0.01	1.21	0.0800	0.0150	-0.0004	0.0006	0.0008	0.0024	0.0024
576	0.810	630.841	-0.01	2.37	0.1901	0.0109	0.0001	0.0006	0.0009	0.0024	0.0024
577	0.810	630.937	-0.01	3.53	0.2926	0.0047	0.0002	0.0006	0.0013	0.0024	0.0024
578	0.811	631.151	-0.00	4.70	0.4050	-0.0026	0.0002	0.0005	0.0007	0.0024	0.0024
579	0.810	630.237	-0.01	5.96	0.5225	-0.0087	0.0003	0.0007	0.0011	0.0024	0.0023
580	0.811	631.411	-0.00	7.16	0.6142	-0.0070	-0.0004	0.0002	0.0020	0.0023	0.0023
581	0.811	631.726	-0.00	8.26	0.6657	-0.0011	-0.0012	0.0001	0.0017	0.0023	0.0023
582	0.812	631.975	-0.00	9.37	0.7012	0.0050	-0.0004	0.0002	0.0014	0.0023	0.0023
583	0.811	631.566	-0.00	10.46	0.7297	0.0087	0.0001	0.0003	0.0015	0.0023	0.0022
584	0.814	633.761	-0.00	11.58	0.7744	0.0118	0.0002	0.0004	0.0007	0.0023	0.0022
585	0.813	633.008	-0.00	12.67	0.8066	0.0132	0.0002	0.0003	0.0013	0.0022	0.0022
586	0.813	633.661	-0.00	13.78	0.8317	0.0145	-0.0001	0.0004	0.0012	0.0022	0.0022
587	0.815	634.699	-0.01	15.99	0.9022	0.0154	0.0000	0.0004	0.0015	0.0021	0.0021
588	0.815	634.905	-0.01	18.08	0.9722	0.0151	0.0001	0.0005	0.0010	0.0019	0.0019
589	0.815	634.901	-0.00	20.19	1.0571	0.0147	0.0004	0.0007	-0.0010	0.0016	0.0017
590	0.819	638.564	-0.00	22.24	1.1421	0.0128	0.0004	0.0004	-0.0004	0.0012	0.0012
591	0.820	639.864	-0.00	23.24	1.1767	0.0116	-0.0002	0.0006	-0.0035	0.0009	0.0009
592	0.812	631.631	-0.00	0.08	-0.0239	0.0159	0.0000	0.0007	-0.0001	0.0024	0.0024
593	0.810	630.215	-0.01	0.09	-0.0216	0.0160	-0.0005	0.0006	0.0003	0.0024	0.0024

TEST= 768 RUN= 25

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	CDC
573	0.812	632.386	-0.01	-1.07	-0.1246	0.0163	0.0019	-0.0007	0.0006	-7.655	0.0024	0.0024
574	0.810	630.827	-0.01	0.05	-0.0198	0.0159	0.0070	-0.0005	0.0006	-1.242	0.0024	0.0024
575	0.811	631.332	-0.01	1.21	0.0796	0.0167	0.0115	-0.0004	0.0006	4.769	0.0024	0.0024
576	0.810	630.841	-0.01	2.37	0.1895	0.0188	0.0152	0.0001	0.0006	10.105	0.0024	0.0024
577	0.810	630.937	-0.01	3.53	0.2918	0.0227	0.0194	0.0003	0.0006	12.843	0.0024	0.0024
578	0.811	631.151	-0.00	4.70	0.4039	0.0306	0.0240	0.0003	0.0004	13.190	0.0024	0.0024
579	0.810	630.237	-0.01	5.96	0.5205	0.0456	0.0278	0.0004	0.0006	11.415	0.0024	0.0023
580	0.811	631.411	-0.00	7.16	0.6102	0.0696	0.0460	-0.0004	0.0002	8.773	0.0023	0.0023
581	0.811	631.726	-0.00	8.26	0.6589	0.0946	0.0741	-0.0012	0.0003	6.968	0.0023	0.0023
582	0.812	631.975	-0.00	9.37	0.6911	0.1191	0.0964	-0.0004	0.0003	5.803	0.0023	0.0023
583	0.811	631.566	-0.00	10.46	0.7160	0.1410	0.1121	0.0001	0.0003	5.079	0.0022	0.0022
584	0.814	633.761	-0.00	11.58	0.7563	0.1670	0.1281	0.0002	0.0004	4.528	0.0022	0.0022
585	0.813	633.008	-0.00	12.67	0.7841	0.1898	0.1429	0.0003	0.0003	4.131	0.0022	0.0022
586	0.813	633.661	-0.00	13.78	0.8043	0.2122	0.1515	-0.0000	0.0004	3.790	0.0021	0.0021
587	0.815	634.699	-0.01	15.99	0.8631	0.2634	0.1645	0.0002	0.0004	3.276	0.0020	0.0020
588	0.815	634.905	-0.01	18.08	0.9195	0.3162	0.1708	0.0002	0.0004	2.908	0.0018	0.0018
589	0.815	634.901	-0.00	20.19	0.9871	0.3786	0.1734	0.0007	0.0005	2.607	0.0015	0.0016
590	0.819	638.564	-0.00	22.24	1.0523	0.4440	0.1823	0.0005	0.0003	2.370	0.0011	0.0011
591	0.820	639.864	-0.00	23.24	1.0712	0.4725	0.1873	0.0001	0.0007	2.267	0.0009	0.0008
592	0.812	631.631	-0.00	0.08	-0.0239	0.0159	0.0072	0.0000	0.0007	-1.504	0.0024	0.0024
593	0.810	630.215	-0.01	0.09	-0.0216	0.0159	0.0068	-0.0005	0.0006	-1.357	0.0024	0.0024

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 768 RUN= 26

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	CDC
594	0.758	580.514	-0.01	-1.06	-0.1218	0.0165	0.0007	-0.0004	0.0007	-7.393	0.0023	0.0023
595	0.757	579.478	-0.01	0.05	-0.0234	0.0161	0.0067	-0.0004	0.0007	-1.451	0.0023	0.0023
596	0.759	582.091	-0.01	1.19	0.0751	0.0169	0.0127	-0.0001	0.0006	4.440	0.0023	0.0023
597	0.758	580.612	-0.00	2.33	0.1729	0.0187	0.0165	-0.0001	0.0006	9.258	0.0023	0.0023
598	0.760	582.345	-0.00	3.48	0.2764	0.0223	0.0223	-0.0001	0.0005	12.377	0.0023	0.0023
599	0.756	579.343	-0.00	4.65	0.3718	0.0292	0.0279	0.0001	0.0005	12.723	0.0023	0.0023
600	0.758	580.395	-0.00	5.80	0.4728	0.0460	0.0350	0.0000	0.0002	10.283	0.0023	0.0022
601	0.757	580.212	-0.00	6.97	0.5475	0.0689	0.0571	-0.0001	0.0002	7.947	0.0022	0.0022
602	0.758	580.561	-0.00	8.14	0.6025	0.0917	0.0814	-0.0003	0.0003	6.568	0.0022	0.0022
603	0.758	580.815	-0.00	9.23	0.6528	0.1134	0.1030	0.0000	0.0003	5.757	0.0022	0.0022
604	0.760	582.581	-0.00	10.35	0.6938	0.1359	0.1214	0.0000	0.0003	5.106	0.0022	0.0021
605	0.763	585.415	-0.00	11.49	0.7268	0.1590	0.1360	0.0000	0.0003	4.570	0.0021	0.0021
606	0.759	582.124	-0.00	12.55	0.7686	0.1840	0.1507	0.0002	0.0001	4.176	0.0021	0.0021
607	0.761	584.002	-0.00	13.71	0.7913	0.2070	0.1607	0.0001	0.0002	3.822	0.0020	0.0020
614	0.759	581.454	-0.01	0.09	-0.0208	0.0160	0.0077	-0.0010	0.0006	-1.299	0.0023	0.0023
615	0.759	582.039	-0.00	15.87	0.8473	0.2563	0.1762	-0.0001	0.0004	3.307	0.0020	0.0020
616	0.762	584.247	-0.00	17.94	0.8995	0.3069	0.1837	0.0000	0.0006	2.931	0.0018	0.0018
617	0.765	586.855	-0.00	20.01	0.9509	0.3624	0.1807	0.0011	0.0000	2.624	0.0015	0.0014
618	0.762	584.399	-0.00	22.01	1.0006	0.4188	0.1896	0.0006	-0.0005	2.389	0.0011	0.0011
619	0.764	586.168	0.00	23.98	1.0663	0.4860	0.2060	-0.0001	0.0005	2.194	0.0007	0.0007
620	0.758	579.860	-0.00	0.09	-0.0224	0.0161	0.0068	-0.0005	0.0006	-1.393	0.0023	0.0023

TEST= 768 RUN= 26

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
594	0.758	580.514	-0.01	-1.06	-0.1221	0.0142	-0.0004	0.0007	0.0008	0.0023	0.0023
595	0.757	579.478	-0.01	0.05	-0.0233	0.0161	-0.0004	0.0007	0.0002	0.0023	0.0023
596	0.759	582.091	-0.01	1.19	0.0754	0.0153	-0.0001	0.0006	0.0006	0.0023	0.0023
597	0.758	580.612	-0.00	2.33	0.1735	0.0116	-0.0001	0.0006	0.0001	0.0023	0.0023
598	0.760	582.345	-0.00	3.48	0.2772	0.0055	-0.0002	0.0005	0.0007	0.0023	0.0023
599	0.756	579.343	-0.00	4.65	0.3729	-0.0010	0.0001	0.0005	0.0012	0.0023	0.0023
600	0.758	580.395	-0.00	5.80	0.4751	-0.0020	0.0000	0.0002	0.0005	0.0023	0.0022
601	0.757	580.212	-0.00	6.97	0.5518	0.0020	-0.0001	0.0002	0.0009	0.0023	0.0022
602	0.758	580.561	-0.00	8.14	0.6094	0.0055	-0.0003	0.0003	0.0015	0.0022	0.0022
603	0.758	580.815	-0.00	9.23	0.6625	0.0072	-0.0000	0.0003	0.0016	0.0022	0.0022
604	0.760	582.581	-0.00	10.35	0.7069	0.0090	-0.0000	0.0003	0.0009	0.0022	0.0022
605	0.763	585.415	-0.00	11.49	0.7440	0.0111	-0.0000	0.0003	0.0003	0.0021	0.0021
606	0.759	582.124	-0.00	12.55	0.7902	0.0126	0.0002	0.0002	0.0007	0.0021	0.0021
607	0.761	584.002	-0.00	13.71	0.8178	0.0136	0.0000	0.0002	0.0005	0.0021	0.0021
614	0.759	581.454	-0.01	0.09	-0.0208	0.0161	-0.0010	0.0006	0.0007	0.0023	0.0023
615	0.759	582.039	-0.00	15.87	0.8851	0.0149	-0.0002	0.0003	0.0012	0.0021	0.0021
616	0.762	584.247	-0.00	17.94	0.9203	0.0149	-0.0001	0.0006	-0.0006	0.0019	0.0019
617	0.765	586.855	-0.00	20.01	1.0175	0.0151	0.0010	0.0004	0.0005	0.0016	0.0015
618	0.762	584.399	-0.00	22.01	1.0846	0.0133	0.0007	-0.0003	0.0027	0.0012	0.0012
619	0.764	586.168	0.00	23.98	1.1718	0.0108	-0.0003	0.0004	-0.0033	0.0008	0.0008
620	0.758	579.860	-0.00	0.09	-0.0223	0.0161	-0.0005	0.0006	0.0003	0.0023	0.0023

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 768 RUN= 27

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLS	CNS	CSF	BODY PRESS.COEFF CAB	CAC
621	0.706	527.974	-0.01	-1.03	-0.1142	0.0145	-0.0008	0.0007	0.0011	0.0022	0.0022
622	0.706	528.663	-0.00	0.09	-0.0213	0.0162	-0.0005	0.0006	0.0003	0.0022	0.0022
623	0.706	527.823	-0.00	1.17	0.0716	0.0156	-0.0003	0.0005	-0.0005	0.0022	0.0022
624	0.707	528.837	-0.00	2.30	0.1666	0.0121	-0.0002	0.0005	0.0002	0.0022	0.0022
625	0.707	529.395	-0.00	3.46	0.2615	0.0065	-0.0003	0.0005	0.0011	0.0022	0.0022
626	0.706	527.953	-0.00	4.58	0.3516	0.0001	0.0003	0.0004	0.0014	0.0022	0.0022
627	0.706	528.541	-0.00	5.76	0.4544	-0.0015	-0.0002	0.0001	0.0011	0.0022	0.0022
628	0.706	528.284	-0.00	6.87	0.5519	0.0032	-0.0012	0.0005	0.0011	0.0022	0.0021
629	0.707	529.590	-0.00	8.06	0.6574	0.0055	-0.0029	0.0003	0.0005	0.0022	0.0021
630	0.707	529.269	-0.00	10.23	0.8963	0.0081	0.0000	0.0003	0.0003	0.0021	0.0021
631	0.706	528.239	-0.00	11.37	0.7355	0.0099	0.0000	0.0002	0.0010	0.0021	0.0021
632	0.708	529.732	-0.00	13.59	0.8083	0.0127	-0.0001	0.0002	0.0000	0.0020	0.0020
633	0.709	531.009	-0.00	15.79	0.8772	0.0141	-0.0002	0.0002	-0.0007	0.0020	0.0020
634	0.709	530.953	-0.00	17.81	0.9579	0.0144	-0.0001	0.0003	0.0000	0.0019	0.0019
635	0.709	531.392	-0.00	19.83	0.9859	0.0158	0.0005	0.0003	0.0009	0.0015	0.0016
636	0.711	533.287	-0.00	21.82	1.0515	0.0139	0.0002	0.0005	0.0005	0.0012	0.0012
637	0.710	531.553	-0.00	23.80	1.1061	0.0113	-0.0002	0.0007	-0.0010	0.0008	0.0007
638	0.706	527.853	-0.00	0.09	-0.0231	0.0163	-0.0002	0.0005	0.0001	0.0022	0.0022

TEST= 768 RUN= 27

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	CDC
621	0.706	527.974	-0.01	-1.03	-0.1139	0.0165	0.0067	-0.0008	0.0007	-0.895	0.0022	0.0022
622	0.706	528.663	-0.00	0.09	-0.0214	0.0162	0.0079	-0.0005	0.0006	-1.321	0.0022	0.0022
623	0.706	527.823	-0.00	1.17	0.0712	0.0170	0.0127	-0.0003	0.0005	4.182	0.0022	0.0022
624	0.707	528.837	-0.00	2.30	0.1654	0.0188	0.0185	-0.0002	0.0005	8.801	0.0022	0.0022
625	0.707	529.395	-0.00	3.46	0.2604	0.0222	0.0236	-0.0003	0.0005	11.719	0.0022	0.0022
626	0.706	527.953	-0.00	4.58	0.3500	0.0262	0.0305	0.0003	0.0004	12.417	0.0022	0.0022
627	0.706	528.541	-0.00	5.76	0.4523	0.0341	0.0360	-0.0002	0.0001	10.266	0.0022	0.0022
628	0.706	528.284	-0.00	6.87	0.5477	0.0408	0.0368	-0.0011	0.0004	7.902	0.0022	0.0021
629	0.707	529.590	-0.00	8.06	0.6607	0.0906	0.0747	-0.0028	0.0007	6.634	0.0021	0.0021
630	0.707	529.269	-0.00	10.23	0.8779	0.1307	0.1231	0.0001	0.0003	5.188	0.0021	0.0020
631	0.706	528.239	-0.00	11.37	0.7170	0.1542	0.1390	0.0001	0.0002	4.649	0.0020	0.0020
632	0.708	529.732	-0.00	13.59	0.7827	0.2022	0.1652	-0.0009	0.0002	3.870	0.0020	0.0020
633	0.709	531.009	-0.00	15.79	0.8403	0.2522	0.1836	-0.0001	0.0003	3.332	0.0019	0.0019
634	0.709	530.953	-0.00	17.81	0.8880	0.3005	0.1918	0.0000	0.0003	2.957	0.0018	0.0018
635	0.709	531.392	-0.00	19.83	0.9221	0.3493	0.1827	0.0005	0.0001	2.640	0.0015	0.0015
636	0.711	533.287	-0.00	21.82	0.9705	0.4036	0.1902	0.0004	0.0004	2.405	0.0011	0.0011
637	0.710	531.553	-0.00	23.80	1.0075	0.4566	0.2018	0.0001	0.0007	2.206	0.0007	0.0006
638	0.706	527.853	-0.00	0.09	-0.0231	0.0163	0.0061	-0.0002	0.0005	-1.420	0.0022	0.0022

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 768 RUN= 28

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
639	0.503	314.968	-0.00	-0.93	-0.0991	0.0160	-0.0006	0.0004	0.0004	0.0020	0.0020
640	0.503	314.351	-0.00	0.10	-0.0189	0.0174	-0.0006	0.0003	0.0004	0.0020	0.0020
641	0.502	313.646	-0.00	1.13	0.0638	0.0168	-0.0003	0.0002	-0.0005	0.0020	0.0020
642	0.502	313.298	-0.00	2.20	0.1445	0.0139	-0.0003	0.0003	-0.0010	0.0020	0.0020
643	0.502	313.744	0.00	3.34	0.2326	0.0091	0.0003	0.0002	-0.0021	0.0020	0.0020
644	0.502	313.751	-0.00	4.37	0.3054	0.0032	-0.0003	-0.0000	0.0002	0.0020	0.0020
645	0.502	313.698	0.00	5.44	0.3916	0.0003	-0.0000	-0.0001	-0.0007	0.0020	0.0020
646	0.502	313.772	0.00	6.59	0.4846	0.0018	0.0002	0.0002	-0.0016	0.0019	0.0019
647	0.502	313.848	-0.00	7.65	0.5521	0.0038	0.0005	0.0001	-0.0003	0.0019	0.0019
648	0.502	313.309	0.00	8.78	0.6094	0.0044	0.0004	0.0001	-0.0011	0.0019	0.0018
649	0.502	313.300	0.00	9.83	0.6616	0.0050	-0.0002	-0.0000	-0.0014	0.0019	0.0019
650	0.502	313.397	0.00	10.96	0.7213	0.0054	-0.0030	0.0001	-0.0019	0.0018	0.0018
651	0.502	314.115	0.00	12.00	0.7730	0.0064	-0.0049	-0.0001	-0.0031	0.0018	0.0018
652	0.502	313.607	0.00	13.15	0.8210	0.0075	-0.0055	0.0001	-0.0032	0.0017	0.0017
653	0.502	313.808	0.00	15.30	0.8704	0.0103	-0.0061	-0.0002	-0.0023	0.0017	0.0017
654	0.504	316.050	0.00	17.30	0.9263	0.0120	-0.0004	0.0001	-0.0029	0.0016	0.0016
655	0.503	315.039	0.00	19.31	0.9834	0.0143	-0.0004	0.0002	-0.0031	0.0015	0.0015
656	0.503	314.624	0.00	21.26	1.0164	0.0154	0.0000	0.0003	-0.0034	0.0012	0.0012
657	0.502	313.304	-0.00	23.15	1.0495	0.0136	-0.0002	0.0007	-0.0035	0.0008	0.0008
658	0.502	313.818	0.00	0.14	-0.0185	0.0179	-0.0004	0.0002	-0.0021	0.0020	0.0020

TEST= 768 RUN= 28

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	CDC
639	0.503	314.968	-0.00	-0.93	-0.0988	0.0176	-0.0005	-0.0006	0.0004	-5.609	0.0020	0.0020
640	0.503	314.351	-0.00	0.10	-0.0189	0.0174	0.0085	-0.0006	0.0003	-1.090	0.0020	0.0020
641	0.502	313.646	-0.00	1.13	0.0634	0.0180	0.0150	-0.0003	0.0002	3.516	0.0020	0.0020
642	0.502	313.298	-0.00	2.20	0.1439	0.0195	0.0201	-0.0003	0.0003	7.390	0.0020	0.0020
643	0.502	313.744	0.00	3.34	0.2316	0.0227	0.0272	0.0003	0.0002	10.225	0.0020	0.0020
644	0.502	313.751	-0.00	4.37	0.3043	0.0264	0.0340	-0.0003	0.0000	11.524	0.0020	0.0020
645	0.502	313.698	0.00	5.44	0.3898	0.0374	0.0436	-0.0000	-0.0001	10.428	0.0020	0.0020
646	0.502	313.772	0.00	6.59	0.4812	0.0574	0.0526	0.0003	0.0002	8.384	0.0019	0.0019
647	0.502	313.848	-0.00	7.65	0.5467	0.0773	0.0694	0.0005	0.0000	7.075	0.0019	0.0019
648	0.502	313.309	0.00	8.78	0.6016	0.0974	0.0917	0.0004	0.0000	6.175	0.0019	0.0018
649	0.502	313.300	0.00	9.83	0.6511	0.1179	0.1160	-0.0002	0.0000	5.521	0.0018	0.0018
650	0.502	313.397	0.00	10.96	0.7071	0.1424	0.1293	-0.0029	0.0006	4.965	0.0018	0.0018
651	0.502	314.115	0.00	12.00	0.7548	0.1670	0.1429	-0.0048	0.0010	4.520	0.0017	0.0017
652	0.502	313.607	0.00	13.15	0.7978	0.1942	0.1616	-0.0054	0.0014	4.109	0.0017	0.0017
653	0.502	313.808	0.00	15.30	0.8368	0.2396	0.1974	-0.0062	-0.0001	3.493	0.0016	0.0016
654	0.504	316.050	0.00	17.30	0.8808	0.2869	0.2145	-0.0004	0.0002	3.070	0.0016	0.0015
655	0.503	315.039	0.00	19.31	0.9233	0.3388	0.2129	-0.0003	0.0003	2.725	0.0014	0.0014
656	0.503	314.624	0.00	21.26	0.9416	0.3829	0.2044	0.0001	0.0003	2.459	0.0011	0.0011
657	0.502	313.304	-0.00	23.15	0.9596	0.4250	0.2094	0.0001	0.0007	2.258	0.0008	0.0008
658	0.502	313.818	0.00	0.14	-0.0185	0.0179	0.0072	-0.0004	0.0002	-1.035	0.0020	0.0020

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 1

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
152	.934	740.428	-.00	.48	.2762	.0334	.0015	.0003	.0007	.0062	0.0000
153	.933	739.665	-.00	1.56	.3723	.0340	.0029	.0001	.0008	.0063	0.0000
154	.932	738.819	-.00	2.75	.4555	.0332	.0038	-.0001	.0014	.0062	0.0000
155	.932	739.171	-.00	3.89	.5520	.0316	.0042	-.0002	.0015	.0062	0.0000
156	.935	741.123	-.00	5.05	.6341	.0294	.0036	-.0003	.0016	.0063	0.0000
157	.935	741.633	-.00	6.21	.7256	.0256	.0022	-.0002	.0023	.0062	0.0000
158	.935	741.853	-.00	7.41	.8082	.0228	.0001	-.0003	.0018	.0062	0.0000
159	.936	742.142	-.00	8.57	.8849	.0192	-.0026	-.0003	.0025	.0061	0.0000
160	.938	743.666	-.00	9.75	.9758	.0177	-.0035	-.0003	.0030	.0059	0.0000
161	.939	744.434	-.00	10.92	1.0432	.0144	-.0051	-.0002	.0020	.0058	0.0000

TEST= 778 RUN= 1

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
152	.934	740.428	-.00	.48	.2759	.0357	-.1515	.0015	.0003	7.739	.0062	33.8823
153	.933	739.665	-.00	1.56	.3712	.0441	-.1451	.0029	-.0000	8.417	.0063	38.4627
154	.932	738.819	-.00	2.75	.4534	.0550	-.1367	.0038	-.0003	8.244	.0062	44.4235
155	.932	739.171	-.00	3.89	.5486	.0689	-.1332	.0042	-.0005	7.959	.0062	46.1800
156	.935	741.123	-.00	5.05	.6291	.0851	-.1343	.0036	-.0006	7.392	.0062	50.3211
157	.935	741.633	-.00	6.21	.7186	.1040	-.1319	.0022	-.0005	6.909	.0062	62.4935
158	.935	741.853	-.00	7.41	.7985	.1257	-.1346	.0001	-.0003	6.301	.0061	95.8736
159	.936	742.142	-.00	8.57	.8722	.1509	-.1261	-.0026	.0001	5.779	.0060	115.7645
160	.938	743.666	-.00	9.75	.9587	.1827	-.1326	-.0035	.0003	5.247	.0058	150.2119
161	.939	744.434	-.00	10.92	1.0216	.2118	-.1272	-.0050	.0008	4.823	.0057	158.8708

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 2

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
174	.863	682.369	-.00	.47	.2967	.0202	.0017	.0004	.0008	.0054	0.0000
175	.862	681.710	-.01	1.69	.4170	.0184	.0017	.0004	.0013	.0054	0.0000
176	.863	682.428	-.00	2.81	.5235	.0168	.0027	.0002	.0020	.0054	0.0000
177	.864	683.866	-.00	4.05	.6146	.0138	.0047	.0001	.0020	.0054	0.0000
178	.863	682.635	-.00	5.17	.7038	.0098	.0056	-.0001	.0027	.0053	0.0000
179	.863	683.244	-.00	6.32	.7786	.0048	.0057	-.0001	.0030	.0053	0.0000
180	.862	681.772	-.00	7.44	.8453	.0005	.0050	-.0001	.0028	.0052	0.0000
181	.863	682.839	-.01	8.60	.8790	-.0018	.0036	.0001	.0030	.0052	0.0000
182	.863	683.198	-.00	9.74	.9370	-.0020	.0039	-.0001	.0019	.0051	0.0000
183	.867	686.037	-.00	10.86	.9664	-.0018	-.0028	-.0004	.0027	.0052	0.0000
185	.868	686.924	-.00	11.98	.9870	-.0006	-.0009	-.0006	.0033	.0052	0.0000
186	.866	685.168	-.00	12.99	.9846	.0015	.0036	-.0005	.0038	.0051	0.0000
187	.869	688.151	-.00	14.09	.9734	.0063	.0001	.0001	.0013	.0051	0.0000
188	.869	688.435	-.00	15.15	1.0043	.0077	.0000	-.0001	.0010	.0049	0.0000
189	.868	686.978	-.00	16.33	1.0508	.0083	-.0002	.0002	.0008	.0047	0.0000
191	.872	689.984	-.00	18.50	1.1158	.0073	.0007	-.0000	.0024	.0045	0.0000

TEST= 778 RUN= 2

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
174	.863	682.369	-.00	.47	.2965	.0226	-.1308	.0017	.0004	13.125	.0054	16.8910
175	.862	681.710	-.01	1.69	.4163	.0307	-.1335	.0017	.0004	13.559	.0054	17.0666
176	.863	682.428	-.00	2.81	.5220	.0425	-.1295	.0027	.0001	12.278	.0054	27.6705
177	.864	683.866	-.00	4.05	.6121	.0572	-.1269	.0046	-.0003	10.705	.0054	40.2195
178	.863	682.635	-.00	5.17	.7001	.0731	-.1205	.0056	-.0006	9.577	.0053	48.6897
179	.863	683.244	-.00	6.32	.7734	.0905	-.1103	.0057	-.0007	8.544	.0053	66.5092
180	.862	681.772	-.00	7.44	.8381	.1099	-.0944	.0050	-.0008	7.623	.0052	107.6696
181	.863	682.839	-.01	8.60	.8694	.1296	-.0670	.0036	-.0005	6.709	.0051	179.5767
182	.863	683.198	-.00	9.74	.9238	.1565	-.0437	.0038	-.0007	5.903	.0050	220.9885
183	.867	686.037	-.00	10.86	.9494	.1803	-.0201	-.0028	.0002	5.266	.0051	300.4239
185	.868	686.924	-.00	11.98	.9656	.2042	.0129	-.0011	-.0004	4.729	.0051	201.4120
186	.866	685.168	-.00	12.99	.9591	.2228	.0358	.0034	-.0013	4.304	.0049	166.7767
187	.869	688.151	-.00	14.09	.9426	.2430	.0512	.0001	.0000	3.879	.0049	134.5880
188	.869	688.435	-.00	15.15	.9674	.2699	.0576	-.0000	-.0001	3.584	.0048	145.5057
189	.868	686.978	-.00	16.33	1.0061	.3035	.0657	-.0001	.0002	3.315	.0045	146.5712
191	.872	689.984	-.00	18.50	1.0559	.3610	.0803	.0006	-.0002	2.925	.0042	146.3202

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 3

					BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
192	.805	630.597	-.01	.33	.2705	.0178	.0016	.0005	.0017	.0050	0.0000
193	.806	631.524	-.01	1.54	.3865	.0138	.0013	.0005	.0025	.0050	0.0000
194	.808	632.958	-.01	2.71	.5002	.0071	.0011	.0004	.0022	.0050	0.0000
195	.807	632.163	-.01	3.90	.6226	-.0008	.0007	.0002	.0031	.0050	0.0000
196	.805	629.908	-.00	5.13	.7200	-.0101	.0006	-.0001	.0026	.0050	0.0000
197	.807	632.049	-.00	6.26	.8183	-.0144	.0018	-.0005	.0043	.0049	0.0000
198	.807	631.865	-.00	7.40	.8652	-.0189	-.0008	-.0004	.0035	.0049	0.0000
199	.806	630.818	-.00	8.53	.9142	-.0211	-.0015	-.0004	.0040	.0049	0.0000
200	.809	633.715	-.00	9.68	.9646	-.0172	-.0002	-.0002	.0032	.0049	0.0000
201	.807	632.373	-.00	10.72	.9866	-.0143	.0042	-.0002	.0040	.0048	0.0000
202	.809	634.209	-.00	11.87	.9875	-.0100	.0106	-.0006	.0057	.0047	0.0000
203	.811	635.706	-.01	12.86	1.0054	-.0027	.0154	-.0002	.0051	.0047	0.0000
204	.810	634.798	-.00	13.97	.9622	.0024	.0005	-.0001	.0032	.0047	0.0000
205	.810	634.708	-.00	16.15	1.0084	.0047	.0010	-.0002	.0032	.0047	0.0000
206	.811	636.043	-.01	18.29	1.0761	.0047	.0016	-.0003	.0052	.0041	0.0000

TEST= 778 RUN= 3

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
192	.805	630.597	-.01	.33	.2703	.0194	-.1163	.0016	.0005	13.946	.0050	14.0800
193	.806	631.524	-.01	1.54	.3860	.0241	-.1164	.0013	.0004	15.999	.0050	13.3772
194	.808	632.958	-.01	2.71	.4993	.0308	-.1186	.0011	.0004	16.234	.0050	15.3098
195	.807	632.163	-.01	3.90	.6212	.0415	-.1164	.0007	.0002	14.955	.0050	15.8117
196	.805	629.908	-.00	5.13	.7180	.0543	-.1136	.0006	-.0002	13.218	.0050	23.1529
197	.807	632.049	-.00	6.26	.8150	.0750	-.1041	.0017	-.0007	10.870	.0049	44.9254
198	.807	631.865	-.00	7.40	.8604	.0928	-.0815	-.0009	-.0003	9.276	.0048	90.4776
199	.806	630.818	-.00	8.53	.9072	.1147	-.0477	-.0015	-.0002	7.911	.0048	124.7361
200	.809	633.715	-.00	9.68	.9537	.1453	-.0260	-.0003	-.0002	6.565	.0048	260.8944
201	.807	632.373	-.00	10.72	.9721	.1695	.0047	.0041	-.0010	5.735	.0047	238.3061
202	.809	634.209	-.00	11.87	.9684	.1933	.0224	.0102	-.0027	5.011	.0046	146.4472
203	.811	635.706	-.01	12.86	.9808	.2212	.0420	.0149	-.0037	4.434	.0046	174.6825
204	.810	634.798	-.00	13.97	.9331	.2345	.0725	.0005	-.0002	3.979	.0046	120.2822
205	.810	634.708	-.00	16.15	.9673	.2851	.0926	.0009	-.0005	3.393	.0045	114.8233
206	.811	636.043	-.01	18.29	1.0203	.3422	.1073	.0015	-.0008	2.982	.0039	109.5528

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 4

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
285	.752	578.353	-.00	.44	.2522	.0172	.0016	.0005	.0004	.0047	0.0000
286	.753	579.566	-.00	1.65	.3626	.0133	.0013	.0005	.0001	.0047	0.0000
287	.753	579.495	-.00	2.79	.4615	.0066	.0008	.0004	.0013	.0047	0.0000
288	.752	578.544	-.00	3.99	.5636	-.0028	.0008	.0003	.0019	.0047	0.0000
289	.753	579.257	-.00	5.15	.6590	-.0137	.0021	.0005	.0002	.0047	0.0000
290	.754	580.443	-.01	6.32	.7506	-.0254	.0022	.0004	.0019	.0047	0.0000
291	.755	581.024	-.00	7.48	.8442	-.0345	.0006	-.0003	.0026	.0046	0.0000
292	.755	581.509	-.00	8.63	.8977	-.0365	.0001	-.0004	.0032	.0046	0.0000
293	.755	581.565	-.00	9.74	.9378	-.0321	.0007	.0003	.0015	.0046	0.0000
294	.754	580.246	-.00	10.82	.9527	-.0269	.0000	-.0002	.0014	.0045	0.0000
295	.753	579.576	-.00	11.89	.9633	-.0183	.0042	-.0001	.0031	.0045	0.0000
296	.755	581.132	-.00	12.94	.9544	-.0104	.0003	.0003	.0021	.0045	0.0000
297	.757	582.980	-.00	13.98	.9645	-.0046	.0007	-.0000	.0024	.0044	0.0000
298	.757	583.130	-.00	16.17	1.0004	.0007	.0010	-.0002	.0027	.0043	0.0000
299	.757	583.829	-.00	18.31	1.0592	.0010	.0014	.0000	.0027	.0040	0.0000
300	.759	584.903	-.00	20.35	1.1207	.0043	.0010	-.0002	.0039	.0034	0.0000
301	.758	584.645	-.01	22.50	1.1883	.0040	.0009	.0000	.0035	.0026	0.0000
302	.760	585.838	-.00	24.55	1.2666	.0022	.0009	-.0002	.0038	.0019	0.0000
303	.752	578.814	-.00	.39	.2619	.0167	.0015	.0005	.0005	.0047	0.0000

TEST= 778 RUN= 4

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
285	.752	578.353	-.00	.44	.2521	.0191	-.1066	.0016	.0005	13.205	.0047	9.3867
286	.753	579.566	-.00	1.65	.3621	.0237	-.1055	.0013	.0004	15.272	.0047	10.0894
287	.753	579.495	-.00	2.79	.4607	.0291	-.1047	.0008	.0003	15.846	.0047	11.2690
288	.752	578.544	-.00	3.99	.5624	.0364	-.1016	.0008	.0003	15.450	.0047	11.6706
289	.753	579.257	-.00	5.15	.6576	.0455	-.0952	.0021	.0003	14.453	.0047	14.9584
290	.754	580.443	-.01	6.32	.7489	.0573	-.0862	.0022	.0001	13.059	.0047	14.8580
291	.755	581.024	-.00	7.48	.8415	.0758	-.0773	.0006	-.0003	11.109	.0046	28.1098
292	.755	581.509	-.00	8.63	.8930	.0986	-.0487	.0001	-.0004	9.054	.0045	88.8462
293	.755	581.565	-.00	9.74	.9297	.1270	-.0118	.0007	.0002	7.319	.0045	210.5705
294	.754	580.246	-.00	10.82	.9408	.1524	.0129	-.0000	-.0002	6.172	.0044	184.9708
295	.753	579.576	-.00	11.89	.9464	.1806	.0387	.0041	-.0009	5.240	.0044	152.0927
296	.755	581.132	-.00	12.94	.9325	.2036	.0663	.0003	.0002	4.581	.0043	150.8378
297	.757	582.980	-.00	13.98	.9370	.2285	.0860	.0007	-.0002	4.100	.0043	105.9127
298	.757	583.130	-.00	16.17	.9606	.2793	.1056	.0009	-.0004	3.439	.0042	104.4068
299	.757	583.829	-.00	18.31	1.0053	.3338	.1235	.0013	-.0004	3.012	.0038	94.1167
300	.759	584.903	-.00	20.35	1.0492	.3938	.1258	.0009	-.0005	2.664	.0032	107.1676
301	.758	584.645	-.01	22.50	1.0963	.4584	.1338	.0009	-.0003	2.392	.0024	117.8351
302	.760	585.838	-.00	24.55	1.1511	.5284	.1521	.0007	-.0006	2.179	.0017	136.0939
303	.752	578.814	-.00	.39	.2618	.0185	-.1082	.0015	.0005	14.169	.0047	10.2651

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 5 06/19/67.

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
325	.702	526.501	-.01	.43	.2389	.0165	.0014	.0006	.0018	.0046	-.0000
326	.702	526.665	-.01	1.51	.3457	.0129	.0015	.0006	.0012	.0046	-.0000
327	.700	524.808	-.00	2.69	.4409	.0065	.0012	.0005	.0010	.0046	-.0000
328	.701	525.522	-.00	3.82	.5302	-.0029	.0012	.0004	.0004	.0046	-.0000
329	.702	526.714	-.01	4.88	.6184	-.0128	.0019	.0005	.0019	.0046	-.0000
330	.702	527.154	-.00	6.05	.7008	-.0249	.0023	.0005	.0008	.0045	-.0000
331	.702	526.644	-.00	7.16	.7710	-.0379	.0018	.0005	.0007	.0045	-.0000
332	.702	527.275	-.00	8.33	.8669	-.0436	-.0006	-.0006	.0034	.0045	-.0000
333	.703	528.098	.00	9.37	.8812	-.0356	-.0040	-.0009	.0038	.0044	-.0000
334	.704	529.011	-.00	10.47	.9097	-.0298	-.0022	-.0002	.0025	.0044	-.0000
335	.701	526.559	-.01	11.55	.9297	-.0223	.0005	.0005	.0020	.0044	-.0000
336	.703	528.505	-.00	12.64	.9526	-.0161	.0021	.0004	.0011	.0043	-.0000
337	.701	526.489	-.00	13.73	.9610	-.0110	.0012	-.0001	.0027	.0043	-.0000
338	.705	529.883	-.00	15.87	.9989	-.0035	.0011	-.0002	.0021	.0042	-.0000
339	.702	526.905	-.01	17.98	1.0335	-.0018	.0013	.0000	.0041	.0040	-.0000
340	.702	527.431	-.00	20.07	1.0926	.0019	.0010	.0002	.0026	.0034	-.0000
341	.706	530.581	-.00	22.07	1.1530	.0034	.0013	-.0001	.0033	.0028	-.0000
342	.702	526.671	-.00	.18	.2480	.0164	.0019	.0005	.0014	.0045	-.0000
343	.701	525.412	-.01	.14	.2413	.0163	.0011	.0006	.0020	.0045	-.0000
344	.702	526.427	-.01	.15	.2434	.0164	.0015	.0006	.0017	.0045	-.0000

TEST= 778 RUN= 5

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	L/D	STAB.PRESS.COEFF CDB	PB-1
325	.702	526.501	-.01	.43	.2387	.0163	-.1040	.0014	.0005	13.072	.0046	7.0149
326	.702	526.665	-.01	1.51	.3452	.0220	-.0985	.0015	.0005	15.662	.0046	7.5545
327	.700	524.808	-.00	2.69	.4401	.0272	-.0972	.0012	.0004	16.204	.0046	8.2447
328	.701	525.522	-.00	3.82	.5292	.0325	-.0951	.0013	.0003	16.273	.0046	8.4330
329	.702	526.714	-.01	4.88	.6172	.0389	-.0843	.0019	.0003	15.475	.0046	8.4079
330	.702	527.154	-.00	6.05	.6995	.0491	-.0735	.0024	.0002	14.236	.0045	9.9639
331	.702	526.644	-.00	7.16	.7697	.0586	-.0659	.0018	.0003	13.143	.0045	10.4659
332	.702	527.275	-.00	8.33	.8640	.0825	-.0503	-.0007	-.0005	10.478	.0045	56.9720
333	.703	528.098	-.00	9.37	.8752	.1043	-.0049	-.0041	-.0003	8.085	.0044	103.2774
334	.704	529.011	-.00	10.47	.8999	.1361	.0241	-.0022	.0002	6.612	.0043	119.2145
335	.701	526.559	-.01	11.55	.9153	.1643	.0508	.0006	.0004	5.570	.0043	127.2458
336	.703	528.505	-.00	12.64	.9330	.1927	.0738	.0022	-.0000	4.841	.0042	125.7400
337	.701	526.489	-.00	13.73	.9362	.2173	.0957	.0012	-.0004	4.307	.0041	100.1402
338	.705	529.883	-.00	15.87	.9618	.2697	.1161	.0010	-.0005	3.566	.0040	94.8697
339	.702	526.905	-.01	17.98	.9836	.3173	.1331	.0012	-.0004	3.100	.0038	88.0933
340	.702	527.431	-.00	20.07	1.0256	.3766	.1362	.0010	-.0002	2.723	.0032	92.6109
341	.706	530.581	-.00	22.07	1.0672	.4364	.1371	.0011	-.0006	2.446	.0026	106.1637
342	.702	526.671	-.00	.18	.2480	.0172	-.1030	.0019	.0005	14.443	.0045	8.1067
343	.701	525.412	-.01	.14	.2412	.0169	-.1009	.0011	.0006	14.247	.0045	8.1067
344	.702	526.427	-.01	.15	.2433	.0170	-.1027	.0015	.0006	14.312	.0045	7.3663

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 6

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
358	.497	310.221	-.00	.22	.2125	.0167	.0018	.0005	-.0001	.0041	0.0000
359	.498	311.021	-.00	1.29	.3073	.0137	.0014	.0005	-.0004	.0041	0.0000
360	.497	309.871	-.00	2.34	.3822	.0081	.0011	.0004	-.0005	.0041	0.0000
361	.498	311.025	-.00	3.43	.4728	.0006	.0012	.0004	-.0011	.0041	0.0000
362	.496	309.790	.00	4.47	.5301	-.0080	.0016	.0003	-.0019	.0041	0.0000
363	.497	310.331	.00	5.56	.6137	-.0183	.0015	.0004	-.0025	.0041	0.0000
364	.496	309.717	.00	6.62	.6741	-.0309	.0012	.0003	-.0027	.0040	0.0000
365	.497	310.341	.00	6.63	.6841	-.0305	.0016	.0003	-.0031	.0041	0.0000
366	.497	310.089	.00	7.71	.7578	-.0437	.0006	.0004	-.0029	.0040	0.0000
367	.498	311.752	.00	8.83	.8129	-.0469	.0003	-.0005	-.0008	.0040	0.0000
368	.501	314.882	.00	9.92	.8726	-.0414	-.0004	-.0004	.0009	.0039	0.0000
369	.500	313.683	.00	11.02	.8974	-.0336	.0009	-.0002	-.0013	.0039	0.0000
370	.500	313.722	.00	12.07	.9297	-.0266	.0019	-.0001	-.0009	.0039	0.0000
371	.500	313.493	.00	13.21	.9469	-.0219	.0013	-.0002	-.0011	.0038	0.0000
372	.499	311.974	.00	15.38	1.0078	-.0139	.0015	-.0001	-.0002	.0037	0.0000
373	.498	310.788	.00	17.48	1.0536	-.0092	.0009	-.0001	-.0005	.0035	0.0000
374	.500	313.259	.00	19.54	1.1024	-.0052	.0013	.0002	-.0015	.0033	0.0000
375	.504	317.007	.00	21.60	1.1608	-.0011	.0012	-.0002	.0004	.0027	0.0000
377	.497	309.590	.00	.25	.2164	.0171	.0013	.0004	-.0019	.0041	0.0000

TEST= 778 RUN= 6

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
358	.497	310.221	-.00	.22	.2124	.0175	-.0926	.0019	.0005	12.137	.0041	4.4047
359	.498	311.021	-.00	1.29	.3069	.0206	-.0845	.0015	.0005	14.900	.0041	4.5929
360	.497	309.871	-.00	2.34	.3815	.0237	-.0836	.0012	.0004	16.109	.0041	4.2792
361	.498	311.025	-.00	3.43	.4719	.0289	-.0759	.0012	.0003	16.313	.0041	4.7059
362	.496	309.790	.00	4.47	.5291	.0333	-.0689	.0016	.0002	15.869	.0040	4.0282
363	.497	310.331	.00	5.56	.6126	.0412	-.0596	.0015	.0003	14.854	.0040	4.4172
364	.496	309.717	.00	6.62	.6731	.0470	-.0546	.0012	.0001	14.312	.0040	4.4298
365	.497	310.341	.00	6.63	.6831	.0487	-.0532	.0017	.0001	14.032	.0040	4.5176
366	.497	310.089	.00	7.71	.7568	.0584	-.0398	.0006	.0003	12.951	.0040	5.2706
367	.498	311.752	.00	8.83	.8104	.0785	-.0303	.0002	-.0005	10.327	.0039	25.2235
368	.501	314.882	.00	9.92	.8667	.1095	-.0002	-.0004	-.0003	7.914	.0039	70.5248
369	.500	313.683	.00	11.02	.8873	.1386	.0352	.0009	-.0004	6.404	.0038	90.6031
370	.500	313.722	.00	12.07	.9147	.1684	.0665	.0019	-.0005	5.432	.0038	91.3560
371	.500	313.493	.00	13.21	.9269	.1951	.0893	.0013	-.0005	4.751	.0037	88.8462
372	.499	311.974	.00	15.38	.9755	.2538	.1273	.0014	-.0005	3.843	.0035	92.6109
373	.498	310.788	.00	17.48	1.0077	.3078	.1493	.0008	-.0004	3.274	.0033	86.3364
374	.500	313.259	.00	19.54	1.0407	.3639	.1646	.0013	-.0002	2.860	.0031	84.0776
375	.504	317.007	.00	21.60	1.0796	.4264	.1663	.0010	-.0006	2.532	.0025	83.5757
377	.497	309.590	.00	.25	.2163	.0181	-.0911	.0013	.0004	11.982	.0041	4.5051

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 7

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
378	.292	119.300	.00	.17	.2165	.0185	.0020	-.0000	-.0065	.0038	0.0000
379	.292	119.204	.00	1.16	.3116	.0164	.0020	.0003	-.0072	.0038	0.0000
380	.292	119.301	.00	2.19	.3580	.0111	.0015	.0000	-.0070	.0038	0.0000
381	.292	119.304	.00	4.19	.5196	-.0035	.0018	.0001	-.0084	.0038	0.0000
382	.290	117.959	.00	6.31	.6298	-.0230	.0013	-.0000	-.0089	.0037	0.0000
383	.292	119.418	.00	8.36	.7662	-.0477	.0006	-.0009	-.0094	.0037	0.0000
384	.290	117.988	.00	9.42	.8243	-.0533	.0001	-.0023	-.0033	.0038	0.0000
385	.291	118.894	.00	10.49	.8934	-.0515	-.0017	-.0020	-.0026	.0036	0.0000
386	.292	119.612	.00	11.57	.9065	-.0421	-.0011	-.0020	-.0041	.0036	0.0000
387	.290	118.402	.00	12.69	.9718	-.0369	-.0011	-.0014	-.0054	.0036	0.0000
388	.291	119.159	.00	14.83	1.0001	-.0254	.0005	-.0009	-.0081	.0034	0.0000
389	.291	119.152	.00	16.90	1.0726	-.0165	.0004	-.0002	-.0094	.0031	0.0000
390	.292	119.718	.00	18.96	1.1210	-.0117	.0004	-.0000	-.0103	.0030	0.0000
391	.291	118.926	.00	20.94	1.1641	-.0077	.0009	.0000	-.0111	.0026	0.0000
392	.291	119.012	.00	.22	.2073	.0194	.0020	-.0000	-.0065	.0039	0.0000

TEST= 778 RUN= 7

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
378	.292	119.300	.00	.17	.2164	.0192	-.0805	.0020	-.0000	11.293	.0038	1.3929
379	.292	119.204	.00	1.16	.3112	.0227	-.0705	.0020	.0002	13.707	.0038	1.5310
380	.292	119.301	.00	2.19	.3573	.0248	-.0692	.0015	-.0000	14.421	.0038	1.5812
381	.292	119.304	.00	4.19	.5185	.0345	-.0592	.0018	-.0001	15.041	.0038	1.5812
382	.290	117.959	.00	6.31	.6285	.0463	-.0457	.0013	-.0002	13.563	.0037	1.3051
383	.292	119.418	.00	8.36	.7650	.0643	-.0222	.0004	-.0009	11.907	.0037	2.5098
384	.290	117.988	.00	9.42	.8219	.0823	-.0170	-.0003	-.0023	9.985	.0037	13.7286
385	.291	118.894	.00	10.49	.8878	.1121	.0019	-.0021	-.0017	7.918	.0035	28.4235
386	.292	119.612	.00	11.57	.8965	.1405	.0351	-.0014	-.0017	6.382	.0035	40.0313
387	.290	118.402	.00	12.69	.9561	.1776	.0729	-.0014	-.0011	5.384	.0035	42.1019
388	.291	119.159	.00	14.83	.9733	.2314	.1168	.0002	-.0010	4.206	.0032	46.0548
389	.291	119.152	.00	16.90	1.0311	.2961	.1515	.0004	-.0003	3.483	.0030	48.5642
390	.292	119.718	.00	18.96	1.0640	.3531	.1763	.0004	-.0002	3.014	.0028	43.1682
391	.291	118.926	.00	20.94	1.0899	.4089	.1837	.0008	-.0003	2.666	.0024	38.5251
392	.291	119.012	.00	.22	.2072	.0202	-.0818	.0020	-.0000	10.246	.0039	1.6816

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 8

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
412	.932	735.784	-.00	.39	.2749	.0322	.0015	.0001	.0007	.0064	0.0000
413	.934	737.330	-.00	1.55	.3635	.0329	-.0002	.0000	.0004	.0064	0.0000
414	.932	735.952	-.00	2.64	.4501	.0329	-.0021	.0000	.0005	.0063	0.0000
415	.933	736.996	-.00	3.80	.5434	.0311	-.0017	.0000	.0007	.0063	0.0000
416	.934	737.329	-.00	4.94	.6261	.0286	-.0011	-.0001	.0017	.0063	0.0000
417	.934	737.867	-.00	6.13	.7197	.0254	-.0029	-.0002	.0019	.0062	0.0000
418	.936	738.965	-.00	7.30	.8033	.0225	-.0046	-.0002	.0020	.0062	0.0000
419	.938	740.346	-.00	8.49	.8799	.0191	-.0044	-.0004	.0023	.0062	0.0000
420	.937	739.734	.00	9.67	.9533	.0157	-.0052	-.0005	.0026	.0060	0.0000
421	.936	738.716	.00	10.82	1.0243	.0124	-.0058	-.0006	.0026	.0058	0.0000
422	.933	736.071	-.00	.29	.2700	.0329	.0013	.0001	.0007	.0063	0.0000

TEST= 778 RUN= 8

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	COB	PB-1
412	.932	735.784	-.00	.39	.2746	.0340	-.1512	.0015	.0001	8.076	.0064	34.6352
413	.934	737.330	-.00	1.55	.3625	.0427	-.1449	-.0001	.0000	8.481	.0064	45.9293
414	.932	735.952	-.00	2.64	.4481	.0536	-.1380	-.0021	.0001	8.357	.0063	47.1215
415	.933	736.996	-.00	3.80	.5401	.0670	-.1381	-.0017	.0001	8.059	.0063	51.8270
416	.934	737.329	-.00	4.94	.6213	.0824	-.1356	-.0011	.0000	7.538	.0063	67.8895
417	.934	737.867	-.00	6.13	.7129	.1022	-.1369	-.0029	.0001	6.977	.0062	78.6816
418	.936	738.965	-.00	7.30	.7940	.1244	-.1373	-.0046	.0004	6.383	.0061	102.5245
419	.938	740.346	-.00	8.49	.8674	.1488	-.1318	-.0044	.0002	5.829	.0061	112.4381
420	.937	739.734	.00	9.67	.9371	.1757	-.1271	-.0052	.0003	5.333	.0059	144.0614
421	.936	738.716	.00	10.82	1.0037	.2045	-.1207	-.0058	.0005	4.909	.0057	160.6259
422	.933	736.071	-.00	.29	.2698	.0343	-.1499	.0013	.0001	7.865	.0063	37.7097

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 9

					BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
436	.862	678.828	-.00	.31	.3106	.0196	.0019	.0002	.0007	.0055	0.0000
437	.862	678.497	-.00	1.47	.4265	.0183	.0022	.0002	.0010	.0054	0.0000
438	.862	678.638	-.00	2.63	.5268	.0163	.0003	.0000	.0009	.0054	0.0000
439	.863	679.494	-.00	3.81	.6190	.0136	-.0002	-.0001	.0009	.0053	0.0000
440	.862	678.554	.00	4.95	.7074	.0086	-.0011	-.0005	.0013	.0053	0.0000
441	.864	680.516	-.00	6.11	.7875	.0061	-.0018	-.0007	.0034	.0053	0.0000
442	.864	680.187	-.00	7.28	.8442	.0013	.0023	-.0006	.0032	.0053	0.0000
443	.864	680.462	.00	8.39	.8960	-.0032	-.0025	-.0006	.0025	.0052	0.0000
444	.865	680.834	-.00	9.53	.9310	-.0016	.0043	-.0003	.0029	.0051	0.0000
445	.864	680.340	.00	10.62	.9632	-.0019	-.0027	-.0007	.0031	.0051	0.0000
446	.865	681.306	.00	11.76	.9846	-.0008	-.0024	-.0006	.0030	.0050	0.0000
447	.867	682.913	.00	12.77	.9716	.0010	-.0036	-.0001	.0002	.0051	0.0000
448	.866	682.326	.00	13.90	.9887	.0028	-.0012	-.0003	.0010	.0050	0.0000
449	.870	685.333	-.00	16.16	1.0446	.0072	-.0003	.0001	-.0003	.0047	0.0000
450	.872	687.271	.00	18.31	1.1111	.0076	-.0010	-.0004	-.0001	.0044	0.0000
451	.863	679.730	-.00	1.51	.4336	.0192	.0011	.0001	.0008	.0055	0.0000

TEST= 778 RUN= 9

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
436	.862	678.828	-.00	.31	.3105	.0213	-.1344	.0019	.0002	14.557	.0055	18.8862
437	.862	678.497	-.00	1.47	.4259	.0292	-.1375	.0022	.0001	14.590	.0054	17.4431
438	.862	678.638	-.00	2.63	.5255	.0404	-.1344	.0003	-.0000	12.995	.0054	28.8627
439	.863	679.494	-.00	3.81	.6167	.0547	-.1331	-.0002	-.0001	11.265	.0053	34.2588
440	.862	678.554	.00	4.95	.7041	.0696	-.1303	-.0012	-.0004	10.110	.0053	53.7093
441	.864	680.516	-.00	6.11	.7824	.0899	-.1240	-.0018	-.0005	8.707	.0053	116.2028
442	.864	680.187	-.00	7.28	.8372	.1082	-.1008	.0023	-.0009	7.737	.0052	139.7947
443	.864	680.462	.00	8.39	.8868	.1276	-.0864	-.0026	-.0002	6.949	.0052	182.4610
444	.865	680.834	-.00	9.53	.9184	.1525	-.0563	.0042	-.0010	6.020	.0050	190.9943
445	.864	680.340	.00	10.62	.9471	.1756	-.0303	-.0028	-.0002	5.392	.0050	307.9533
446	.865	681.306	.00	11.76	.9641	.1999	-.0040	-.0025	-.0001	4.824	.0049	203.6708
447	.867	682.913	.00	12.77	.9474	.2157	.0233	-.0035	.0007	4.392	.0050	140.0472
448	.866	682.326	.00	13.90	.9590	.2402	.0328	-.0013	.0000	3.993	.0048	132.5178
449	.870	685.333	-.00	16.16	1.0013	.2975	.0367	-.0003	.0002	3.365	.0045	144.9413
450	.872	687.271	.00	18.31	1.0525	.3563	.0405	-.0011	-.0001	2.954	.0042	148.7060
451	.863	679.730	-.00	1.51	.4329	.0306	-.1389	.0011	.0000	14.137	.0055	18.1961

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 10

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
465	.805	627.707	-.00	.25	.2740	.0170	.0019	.0002	.0015	.0051	0.0000
466	.805	627.710	-.00	1.45	.3915	.0130	.0020	.0002	.0021	.0051	0.0000
467	.807	628.790	-.00	2.59	.5044	.0063	.0019	.0002	.0017	.0050	0.0000
468	.807	629.334	-.00	3.80	.6224	-.0019	.0010	.0001	.0019	.0050	0.0000
469	.805	627.564	-.00	4.94	.7194	-.0109	.0006	-.0001	.0018	.0050	0.0000
470	.806	628.348	.00	6.12	.8114	-.0157	.0000	-.0006	.0027	.0049	0.0000
471	.806	628.253	.00	7.27	.8660	-.0196	-.0024	-.0007	.0029	.0049	0.0000
472	.808	630.199	-.00	8.37	.9002	-.0177	.0002	-.0005	.0029	.0048	0.0000
473	.809	631.054	-.00	9.51	.9645	-.0154	.0008	-.0001	.0017	.0048	0.0000
474	.809	631.446	-.00	10.60	.9859	-.0139	.0024	-.0001	.0028	.0048	0.0000
475	.807	629.904	.00	11.69	.9919	-.0112	.0058	-.0012	.0060	.0047	0.0000
476	.806	630.372	-.00	12.74	1.0091	-.0052	.0150	-.0006	.0062	.0047	0.0000
477	.809	631.629	-.00	13.80	.9676	-.0007	-.0001	-.0002	.0032	.0047	0.0000
478	.811	633.542	-.00	16.02	1.0096	.0039	.0008	-.0001	.0031	.0045	0.0000
479	.810	632.303	-.00	18.16	1.0669	.0036	.0008	-.0003	.0037	.0041	0.0000
480	.805	627.667	-.01	.28	.2797	.0169	.0023	.0003	.0023	.0050	0.0000
481	.806	628.664	-.01	.29	.2847	.0169	.0020	.0002	.0025	.0050	0.0000

TEST= 778 RUN= 10

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
465	.805	627.707	-.00	.25	.2739	.0182	-.1186	.0019	.0002	15.059	.0051	11.2188
466	.805	627.710	-.00	1.45	.3911	.0229	-.1215	.0020	.0001	17.070	.0050	13.4776
467	.807	628.790	-.00	2.59	.5036	.0291	-.1232	.0019	.0001	17.316	.0050	14.6070
468	.807	629.334	-.00	3.80	.6212	.0393	-.1245	.0010	.0000	15.796	.0050	15.4102
469	.805	627.564	-.00	4.94	.7177	.0511	-.1223	.0006	-.0001	14.037	.0049	23.6549
470	.806	628.348	.00	6.12	.8084	.0708	-.1126	-.0001	-.0006	11.412	.0049	61.2386
471	.806	628.253	.00	7.27	.8615	.0901	-.0907	-.0025	-.0003	9.561	.0048	95.3716
472	.808	630.199	-.00	8.37	.8932	.1136	-.0561	.0001	-.0005	7.860	.0048	117.2067
473	.809	631.054	-.00	9.51	.9538	.1441	-.0415	.0008	-.0002	6.621	.0048	313.9768
474	.809	631.446	-.00	10.60	.9716	.1677	-.0175	.0024	-.0005	5.793	.0047	307.2003
475	.807	629.904	.00	11.69	.9736	.1901	.0058	.0054	-.0024	5.122	.0046	166.4002
476	.806	630.372	-.00	12.74	.9854	.2176	.0228	.0145	-.0039	4.529	.0046	166.7767
477	.809	631.629	-.00	13.80	.9398	.2300	.0531	-.0001	-.0002	4.086	.0045	109.9295
478	.811	633.542	-.00	16.02	.9693	.2825	.0623	.0007	-.0003	3.432	.0043	120.4707
479	.810	632.303	-.00	18.16	1.0127	.3360	.0755	.0006	-.0005	3.014	.0039	101.2707
480	.805	627.667	-.01	.28	.2796	.0183	-.1193	.0023	.0003	15.258	.0050	12.5490
481	.806	628.664	-.01	.29	.2846	.0184	-.1192	.0021	.0002	15.504	.0050	12.1223

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 11

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
495	.753	577.408	-.00	.21	.2592	.0164	.0020	.0002	.0014	.0048	0.0000
496	.754	578.452	-.00	1.35	.3620	.0125	.0017	.0002	.0011	.0048	0.0000
497	.755	579.734	-.00	2.47	.4684	.0060	.0015	.0002	.0008	.0049	0.0000
498	.752	577.042	-.00	3.63	.5702	-.0037	.0014	.0001	.0004	.0048	0.0000
499	.755	579.756	-.00	4.81	.6694	-.0150	.0024	.0002	.0002	.0048	0.0000
500	.754	578.683	-.00	5.94	.7635	-.0265	.0018	-.0001	.0013	.0048	0.0000
501	.754	578.824	.00	7.12	.8499	-.0348	.0002	-.0003	.0008	.0047	0.0000
502	.754	579.113	-.00	8.22	.8820	-.0318	.0040	.0003	.0012	.0047	0.0000
503	.755	579.331	-.00	9.34	.9371	-.0310	-.0013	-.0002	.0012	.0046	0.0000
504	.756	580.356	.00	10.40	.9418	-.0245	.0038	-.0003	.0010	.0046	0.0000
505	.754	579.148	-.00	11.49	.9664	-.0182	.0061	-.0002	.0019	.0045	0.0000
506	.755	580.152	-.00	12.58	.9951	-.0114	.0109	-.0003	.0019	.0045	0.0000
507	.755	579.445	-.00	13.65	.9608	-.0062	-.0003	-.0004	.0027	.0045	0.0000
508	.756	581.033	-.00	15.83	.9954	.0008	.0009	-.0003	.0023	.0043	0.0000
509	.760	584.122	-.00	17.96	1.0485	.0013	.0012	-.0005	.0026	.0041	0.0000
510	.759	583.837	-.00	20.04	1.1074	.0047	.0009	-.0003	.0028	.0035	0.0000
511	.759	583.536	-.00	22.09	1.1624	.0052	.0004	-.0003	.0019	.0027	0.0000
512	.753	578.008	-.00	.21	.2591	.0163	.0024	.0002	.0010	.0048	0.0000

TEST= 778 RUN= 11

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
495	.753	577.408	-.00	.21	.2591	.0174	-.1118	.0020	.0002	14.902	.0048	9.5121
496	.754	578.452	-.00	1.35	.3616	.0210	-.1109	.0017	.0002	17.210	.0048	9.4117
497	.755	579.734	-.00	2.47	.4677	.0261	-.1117	.0015	.0001	17.900	.0049	10.3404
498	.752	577.042	-.00	3.63	.5693	.0324	-.1106	.0014	.0000	17.545	.0048	11.5702
499	.755	579.756	-.00	4.81	.6683	.0411	-.1067	.0025	.0000	16.263	.0048	15.6862
500	.754	578.683	-.00	5.94	.7621	.0526	-.0988	.0018	-.0002	14.478	.0047	15.7615
501	.754	578.824	.00	7.12	.8476	.0708	-.0899	.0002	-.0003	11.966	.0047	31.7490
502	.754	579.113	-.00	8.22	.8774	.0946	-.0505	.0040	-.0003	9.278	.0046	87.9678
503	.755	579.331	-.00	9.34	.9297	.1214	-.0272	-.0013	.0000	7.659	.0046	265.4121
504	.756	580.356	.00	10.40	.9308	.1459	-.0008	.0036	-.0010	6.379	.0045	136.2825
505	.754	579.148	-.00	11.49	.9506	.1746	.0181	.0060	-.0015	5.444	.0044	163.7649
506	.755	580.152	-.00	12.58	.9737	.2057	.0349	.0106	-.0027	4.734	.0044	152.4708
507	.755	579.445	-.00	13.65	.9351	.2207	.0667	-.0004	-.0003	4.237	.0043	98.2589
508	.756	581.033	-.00	15.83	.9575	.2723	.0790	.0008	-.0005	3.516	.0041	99.7648
509	.760	584.122	-.00	17.96	.9970	.3246	.0912	.0010	-.0008	3.071	.0039	89.6001
510	.759	583.837	-.00	20.04	1.0387	.3840	.0854	.0007	-.0006	2.705	.0033	99.0119
511	.759	583.536	-.00	22.09	1.0750	.4420	.0842	.0002	-.0004	2.432	.0025	110.3060
512	.753	578.008	-.00	.21	.2591	.0172	-.1095	.0024	.0002	15.048	.0048	9.3114

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 12

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
528	.702	525.865	-.00	.15	.2487	.0159	.0020	.0003	.0013	.0046	0.0000
529	.702	525.783	-.00	1.20	.3404	.0126	.0017	.0003	.0012	.0046	0.0000
530	.701	524.767	-.00	2.36	.4462	.0061	.0015	.0002	.0008	.0046	0.0000
531	.702	526.491	-.00	3.51	.5386	-.0031	.0014	.0002	.0004	.0046	0.0000
532	.702	526.421	-.00	4.62	.6240	-.0137	.0021	.0002	.0005	.0046	0.0000
533	.702	526.532	-.00	5.75	.7045	-.0252	.0023	.0003	-.0003	.0046	0.0000
534	.702	525.949	-.00	6.92	.7947	-.0385	.0019	.0003	.0009	.0045	0.0000
535	.702	526.197	-.00	8.04	.8618	-.0420	.0017	.0001	.0003	.0045	0.0000
536	.702	526.380	.00	9.11	.8766	-.0352	-.0031	-.0010	.0020	.0044	0.0000
537	.704	528.239	.00	10.20	.8972	-.0273	-.0045	-.0010	.0037	.0044	0.0000
538	.703	527.444	-.00	11.33	.9492	-.0225	.0037	-.0001	.0010	.0043	0.0000
539	.703	527.414	-.00	12.39	.9577	-.0165	.0047	-.0003	.0022	.0043	0.0000
540	.704	528.160	.00	13.51	.9628	-.0109	.0002	-.0004	.0015	.0043	0.0000
541	.703	527.145	-.00	15.69	1.0006	-.0036	.0008	-.0002	.0017	.0042	0.0000
542	.704	528.555	-.00	17.81	1.0537	-.0015	.0012	-.0003	.0021	.0039	0.0000
543	.705	529.580	-.00	19.85	1.0881	.0026	.0012	-.0002	.0020	.0034	0.0000
544	.705	529.772	-.00	21.88	1.1345	.0043	.0005	-.0002	.0012	.0027	0.0000
545	.701	525.400	-.00	.15	.2488	.0160	.0018	.0002	.0015	.0046	0.0000

TEST= 778 RUN= 12

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
528	.702	525.865	-.00	.15	.2486	.0166	-.1047	.0020	.0003	14.975	.0046	7.8055
529	.702	525.783	-.00	1.20	.3400	.0197	-.1048	.0017	.0002	17.220	.0046	7.1780
530	.701	524.767	-.00	2.36	.4456	.0245	-.1037	.0016	.0001	18.200	.0046	8.1318
531	.702	526.491	-.00	3.51	.5378	.0299	-.1017	.0014	.0001	18.006	.0046	8.7090
532	.702	526.421	-.00	4.62	.6230	.0366	-.0923	.0021	.0001	17.007	.0046	8.8345
533	.702	526.532	-.00	5.75	.7035	.0455	-.0843	.0023	.0000	15.449	.0046	10.7670
534	.702	525.949	-.00	6.92	.7936	.0575	-.0792	.0019	.0001	13.795	.0045	10.9176
535	.702	526.197	-.00	8.04	.8592	.0790	-.0622	.0017	-.0001	10.879	.0045	54.9642
536	.702	526.380	.00	9.11	.8711	.1040	-.0196	-.0033	-.0005	8.378	.0044	92.6109
537	.704	528.239	.00	10.20	.8879	.1321	.0129	-.0046	-.0002	6.720	.0043	108.4225
538	.703	527.444	-.00	11.33	.9351	.1644	.0302	.0036	-.0009	5.690	.0043	133.5203
539	.703	527.414	-.00	12.39	.9389	.1894	.0523	.0045	-.0013	4.958	.0042	111.9362
540	.704	528.160	.00	13.51	.9388	.2142	.0740	.0001	-.0004	4.382	.0042	93.6148
541	.703	527.145	-.00	15.69	.9642	.2671	.0928	.0007	-.0004	3.610	.0040	93.3638
542	.704	528.555	-.00	17.81	1.0037	.3208	.1040	.0011	-.0006	3.128	.0038	86.3364
543	.705	529.580	-.00	19.85	1.0225	.3719	.0990	.0011	-.0007	2.750	.0032	89.5991
544	.705	529.772	-.00	21.88	1.0512	.4267	.0937	.0004	-.0004	2.464	.0025	105.4108
545	.701	525.400	-.00	.15	.2487	.0167	-.1048	.0018	.0002	14.892	.0046	7.8557

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 13

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
558	.499	311.782	-.00	.14	.2190	.0153	.0021	.0003	.0020	.0041	0.0000
559	.498	311.165	-.00	1.17	.3076	.0128	.0019	.0004	.0016	.0041	0.0000
560	.498	311.166	-.00	2.25	.3807	.0073	.0012	.0003	.0019	.0041	0.0000
561	.498	311.078	-.00	3.31	.4693	-.0002	.0012	.0002	.0013	.0041	0.0000
562	.498	310.553	-.00	4.38	.5329	-.0090	.0019	.0004	.0003	.0041	0.0000
563	.498	310.561	-.00	5.45	.6060	-.0194	.0012	.0003	.0003	.0041	0.0000
564	.497	310.128	-.00	6.55	.6734	-.0321	.0012	.0003	-.0001	.0041	0.0000
565	.497	309.613	-.00	7.63	.7444	-.0451	.0005	.0002	-.0001	.0040	0.0000
566	.497	310.397	.00	8.74	.8166	-.0482	.0000	-.0006	.0020	.0040	0.0000
567	.498	310.601	.00	9.81	.8702	-.0434	-.0006	-.0006	.0016	.0039	0.0000
568	.497	310.110	-.00	10.93	.8897	-.0350	.0010	-.0002	.0015	.0038	0.0000
569	.500	313.250	-.00	11.99	.9206	-.0280	.0019	-.0001	.0019	.0038	0.0000
570	.496	308.496	.00	13.13	.9482	-.0233	.0015	-.0004	.0017	.0038	0.0000
571	.497	309.993	.00	15.30	1.0076	-.0149	.0012	-.0001	.0007	.0036	0.0000
572	.498	311.283	-.00	17.39	1.0388	-.0100	.0016	-.0003	.0020	.0034	0.0000
573	.499	312.249	-.00	19.45	1.0925	-.0056	.0012	-.0001	.0018	.0032	0.0000
574	.499	312.236	.00	21.51	1.1355	-.0015	.0006	-.0000	.0001	.0026	0.0000
575	.498	310.727	-.00	.26	.2199	.0159	.0021	.0002	-.0003	.0041	0.0000
576	.498	311.346	-.00	.20	.2229	.0160	.0019	.0004	-.0002	.0041	0.0000

TEST= 778 RUN= 13

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
558	.499	311.782	-.00	.14	.2190	.0159	-.0926	.0021	.0003	13.816	.0041	4.5929
559	.498	311.165	-.00	1.17	.3073	.0191	-.0885	.0020	.0003	16.112	.0041	4.2667
560	.498	311.166	-.00	2.25	.3802	.0222	-.0872	.0012	.0002	17.136	.0041	4.4423
561	.498	311.078	-.00	3.31	.4685	.0269	-.0829	.0013	.0001	17.443	.0041	4.4298
562	.498	310.553	-.00	4.38	.5320	.0317	-.0776	.0019	.0002	16.785	.0041	4.0659
563	.498	310.561	-.00	5.45	.6051	.0383	-.0697	.0013	.0002	15.818	.0040	4.3545
564	.497	310.128	-.00	6.55	.6726	.0450	-.0652	.0012	.0002	14.962	.0040	4.2416
565	.497	309.613	-.00	7.63	.7438	.0541	-.0535	.0005	.0002	13.738	.0040	4.8816
566	.497	310.397	.00	8.74	.8144	.0764	-.0425	-.0001	-.0006	10.662	.0039	24.0313
567	.498	310.601	.00	9.81	.8649	.1054	-.0151	-.0007	-.0005	8.202	.0039	62.4935
568	.497	310.110	-.00	10.93	.8802	.1344	.0176	.0010	-.0004	6.552	.0038	84.0776
569	.500	313.250	-.00	11.99	.9063	.1638	.0482	.0019	-.0005	5.533	.0037	80.9404
570	.496	308.496	.00	13.13	.9288	.1927	.0691	.0014	-.0007	4.819	.0037	87.0894
571	.497	309.993	.00	15.30	.9759	.2515	.1034	.0012	-.0005	3.880	.0035	90.7285
572	.498	311.283	-.00	17.39	.9943	.3010	.1211	.0015	-.0008	3.303	.0033	80.3130
573	.499	312.249	-.00	19.45	1.0320	.3585	.1319	.0011	-.0005	2.879	.0030	77.8032
574	.499	312.236	.00	21.51	1.0570	.4149	.1243	.0005	-.0002	2.548	.0024	81.6933
575	.498	310.727	-.00	.26	.2199	.0169	-.0930	.0021	.0002	13.024	.0041	4.3420
576	.498	311.346	-.00	.20	.2229	.0168	-.0910	.0019	.0004	13.285	.0041	4.3043

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 14

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
591	.292	119.872	-.00	.16	.2148	.0144	.0020	.0005	-.0005	.0038	0.0000
592	.292	119.197	-.00	1.16	.3118	.0119	.0026	.0004	-.0016	.0038	0.0000
593	.292	119.873	-.00	2.16	.3563	.0073	.0021	.0005	-.0014	.0038	0.0000
594	.292	119.199	.00	3.20	.4441	.0007	.0021	.0004	-.0018	.0038	0.0000
595	.291	119.102	.00	4.18	.5107	-.0083	.0018	.0001	-.0020	.0038	0.0000
596	.291	119.107	.00	5.26	.5864	-.0168	.0020	.0004	-.0028	.0039	0.0000
597	.291	119.109	.00	6.31	.6330	-.0278	.0011	.0001	-.0024	.0039	0.0000
598	.291	119.113	.00	7.33	.7101	-.0405	.0008	-.0001	-.0027	.0037	0.0000
599	.292	119.217	.00	8.41	.7579	-.0523	.0008	-.0004	-.0030	.0037	0.0000
600	.292	119.238	-.00	9.43	.8341	-.0572	.0001	-.0016	.0091	.0037	0.0000
601	.292	119.270	.00	10.51	.8815	-.0565	-.0015	-.0024	.0038	.0036	0.0000
602	.291	118.639	.00	11.57	.9231	-.0470	-.0011	-.0017	.0083	.0036	0.0000
603	.292	119.454	-.00	12.70	.9440	-.0413	-.0009	-.0012	.0072	.0035	0.0000
604	.292	119.445	-.00	14.86	1.0075	-.0294	.0009	-.0006	.0041	.0034	0.0000
605	.292	119.629	-.00	16.95	1.0690	-.0206	.0011	-.0004	.0028	.0031	0.0000
606	.293	120.180	-.00	20.97	1.1619	-.0114	.0008	-.0001	.0018	.0026	0.0000
607	.294	120.977	.00	9.49	.8415	-.0578	-.0004	-.0016	.0034	.0037	0.0000

TEST= 77a RUN= 14

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
591	.292	119.872	-.00	.16	.2147	.0150	-.0837	.0020	.0005	14.349	.0038	1.4557
592	.292	119.197	-.00	1.16	.3115	.0183	-.0775	.0026	.0003	17.062	.0038	1.5310
593	.292	119.873	-.00	2.16	.3558	.0207	-.0758	.0022	.0004	17.207	.0038	1.4933
594	.292	119.199	.00	3.20	.4433	.0255	-.0742	.0021	.0002	17.387	.0038	1.4306
595	.291	119.102	.00	4.18	.5100	.0289	-.0708	.0018	.0000	17.627	.0038	1.2925
596	.291	119.107	.00	5.26	.5854	.0370	-.0594	.0020	.0002	15.822	.0038	1.3553
597	.291	119.109	.00	6.31	.6322	.0419	-.0546	.0011	-.0000	15.072	.0038	1.4055
598	.291	119.113	.00	7.33	.7095	.0504	-.0460	.0008	-.0002	14.064	.0037	1.3553
599	.292	119.217	.00	8.41	.7574	.0591	-.0406	.0007	-.0005	12.810	.0037	2.3341
600	.292	119.238	-.00	9.43	.8323	.0802	-.0322	-.0002	-.0016	10.382	.0037	14.3059
601	.292	119.270	.00	10.51	.8770	.1053	-.0165	-.0019	-.0021	8.330	.0035	26.7921
602	.291	118.639	.00	11.57	.9138	.1391	.0190	-.0014	-.0015	6.570	.0035	39.7176
603	.292	119.454	-.00	12.70	.9299	.1673	.0459	-.0011	-.0010	5.559	.0034	42.5411
604	.292	119.445	-.00	14.86	.9813	.2299	.0933	.0007	-.0008	4.268	.0032	46.1176
605	.292	119.629	-.00	16.95	1.0286	.2920	.1232	.0009	-.0007	3.523	.0030	44.3607
606	.293	120.180	-.00	20.97	1.0891	.4051	.1484	.0007	-.0004	2.688	.0024	36.5803
607	.294	120.977	.00	9.49	.8395	.0817	-.0330	-.0007	-.0015	10.271	.0036	13.9294

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 15

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
639	.929	731.612	-.00	.11	.1404	.0213	.0005	.0004	-.0000	.0063	0.0000
640	.931	733.140	-.00	1.31	.2598	.0239	-.0000	.0004	-.0008	.0063	0.0000
641	.933	734.777	-.00	2.48	.3670	.0256	-.0015	.0005	-.0000	.0063	0.0000
642	.932	733.893	-.00	3.68	.4583	.0246	-.0013	.0004	-.0008	.0064	0.0000
643	.933	735.033	-.00	4.87	.5502	.0232	-.0010	.0003	.0005	.0066	0.0000
644	.931	733.275	-.00	6.07	.6502	.0209	.0008	.0006	-.0006	.0066	0.0000
645	.934	735.610	-.00	7.30	.7512	.0200	-.0003	.0003	.0010	.0068	0.0000
646	.935	736.425	-.00	8.51	.8354	.0185	.0029	.0004	.0006	.0067	0.0000
647	.936	737.282	-.00	9.71	.9139	.0169	.0027	.0003	.0002	.0060	0.0000
648	.940	740.064	-.00	10.85	.9839	.0168	.0027	.0001	-.0001	.0058	0.0000
649	.933	734.629	-.00	.16	.1354	.0207	.0006	.0004	.0001	.0063	0.0000

TEST= 778 RUN= 15

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
639	.929	731.612	-.00	.11	.1404	.0216	-.0766	.0005	.0004	6.514	.0063	15.9385
640	.931	733.140	-.00	1.31	.2592	.0299	-.0858	-.0000	.0004	8.682	.0063	19.3714
641	.933	734.777	-.00	2.48	.3555	.0410	-.0850	-.0015	.0006	8.679	.0063	24.3369
642	.932	733.893	-.00	3.68	.4558	.0540	-.0830	-.0012	.0005	8.434	.0064	27.5859
643	.933	735.033	-.00	4.87	.5462	.0698	-.0805	-.0010	.0004	7.823	.0065	42.7888
644	.931	733.275	-.00	6.07	.6444	.0896	-.0789	.0008	.0005	7.194	.0066	61.1794
645	.934	735.610	-.00	7.30	.7426	.1154	-.0847	-.0002	.0003	6.437	.0068	113.0418
646	.935	736.425	-.00	8.51	.8234	.1419	-.0775	.0029	-.0000	5.801	.0067	133.1490
647	.936	737.282	-.00	9.71	.8982	.1709	-.0710	.0028	-.0001	5.256	.0059	166.7428
648	.940	740.064	-.00	10.85	.9636	.2018	-.0727	.0026	-.0004	4.775	.0057	199.8461
649	.933	734.629	-.00	.16	.1353	.0211	-.0802	.0006	.0004	6.412	.0063	17.6550

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 16

POINT	MACH	Q	BETA	ALPHA	CNF	BODY AXIS COEFFICIENTS CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
663	.863	677.534	-.01	.02	.1289	.0174	.0013	.0005	.0009	.0055	0.0000
664	.865	679.244	-.01	1.19	.2490	.0152	.0010	.0005	.0017	.0054	0.0000
665	.864	678.719	-.01	2.40	.3787	.0111	.0008	.0004	.0014	.0054	0.0000
666	.863	677.791	-.00	3.62	.5093	.0065	.0011	.0004	.0009	.0053	0.0000
667	.863	678.173	-.01	4.79	.6147	.0029	.0012	.0004	.0014	.0053	0.0000
668	.862	677.355	-.00	5.97	.7024	.0008	-.0011	-.0002	.0017	.0053	0.0000
669	.864	679.087	-.00	7.17	.7768	.0001	-.0023	.0003	.0008	.0053	0.0000
670	.865	679.413	-.00	8.29	.8128	.0026	-.0013	.0002	.0007	.0052	0.0000
671	.866	680.517	-.01	9.42	.8396	.0058	.0024	.0007	.0006	.0051	0.0000
672	.865	679.244	-.01	10.49	.8606	.0070	.0052	.0010	.0004	.0051	0.0000
673	.863	677.616	-.01	3.63	.5112	.0043	.0013	.0003	.0017	.0053	0.0000

TEST= 778 RUN= 16

POINT	MACH	Q	BETA	ALPHA	CL	STABILITY AXIS COEFFICIENTS CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
663	.863	677.534	-.01	.02	.1289	.0174	-.0572	.0013	.0005	7.408	.0055	11.0099
664	.865	679.244	-.01	1.19	.2487	.0204	-.0602	.0010	.0005	12.191	.0054	15.3992
665	.864	678.719	-.01	2.40	.3779	.0269	-.0642	.0008	.0003	14.048	.0054	20.0458
666	.863	677.791	-.00	3.62	.5080	.0387	-.0756	.0011	.0003	13.127	.0053	20.3523
667	.863	678.173	-.01	4.79	.6125	.0542	-.0779	.0012	.0003	11.301	.0053	27.0955
668	.862	677.355	-.00	5.97	.6987	.0739	-.0678	-.0011	-.0001	9.455	.0053	63.3863
669	.864	679.087	-.00	7.17	.7710	.0971	-.0465	-.0023	.0006	7.940	.0052	71.1103
670	.865	679.413	-.00	8.29	.8042	.1199	-.0036	-.0013	.0004	6.707	.0052	73.9302
671	.866	680.517	-.01	9.42	.8276	.1431	.0280	.0024	.0003	5.783	.0050	99.6781
672	.865	679.244	-.01	10.49	.8453	.1635	.0526	.0052	.0000	5.170	.0050	105.5632
673	.863	677.616	-.01	3.63	.5099	.0366	-.0736	.0013	.0002	13.923	.0053	20.9653

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 17

					BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
687	.805	625.876	-.01	.02	.1152	.0168	.0014	.0006	.0008	.0051	0.0000
688	.807	627.454	-.01	1.19	.2232	.0144	.0013	.0006	.0015	.0051	0.0000
689	.805	625.957	-.01	2.36	.3379	.0091	.0013	.0006	.0021	.0050	0.0000
690	.807	628.039	-.01	3.53	.4512	.0008	.0017	.0005	.0011	.0050	0.0000
691	.806	626.323	-.01	4.73	.5632	-.0083	.0019	.0004	.0016	.0050	0.0000
692	.806	627.030	-.00	5.96	.6925	-.0128	.0003	.0001	.0012	.0049	0.0000
693	.808	628.823	-.00	7.14	.7602	-.0137	-.0010	.0000	.0013	.0049	0.0000
694	.806	627.185	-.01	8.24	.7841	-.0090	.0019	.0005	.0009	.0048	0.0000
695	.806	626.899	-.01	9.32	.8124	-.0051	.0044	.0011	.0012	.0048	0.0000
696	.808	628.288	-.01	10.41	.8302	-.0009	.0021	.0006	.0014	.0048	0.0000
697	.810	630.654	-.01	11.51	.8421	.0054	.0009	.0005	.0007	.0047	0.0000
698	.808	628.823	-.01	12.57	.8677	.0083	.0010	.0006	.0014	.0047	0.0000
699	.809	629.727	-.00	13.70	.8856	.0109	.0006	.0006	.0002	.0047	0.0000
700	.812	632.117	-.00	15.98	.9501	.0123	.0006	.0005	.0005	.0045	0.0000
701	.814	634.420	-.01	18.12	1.0076	.0132	.0007	.0007	-.0001	.0041	0.0000
702	.807	627.222	-.01	.05	.1146	.0152	.0010	.0006	.0023	.0051	0.0000

TEST= 778 RUN= 17

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
687	.805	625.876	-.01	.02	.1152	.0168	-.0527	.0014	.0006	6.857	.0051	9.9800
688	.807	627.454	-.01	1.19	.2229	.0191	-.0512	.0013	.0006	11.670	.0050	10.6666
689	.805	625.957	-.01	2.36	.3374	.0230	-.0493	.0013	.0005	14.670	.0050	11.7701
690	.807	628.039	-.01	3.53	.4504	.0286	-.0477	.0018	.0004	15.748	.0050	15.3501
691	.806	626.323	-.01	4.73	.5621	.0382	-.0462	.0019	.0003	14.715	.0049	18.0965
692	.806	627.030	-.00	5.96	.6903	.0592	-.0483	.0003	.0000	11.660	.0049	32.4901
693	.808	628.823	-.00	7.14	.7563	.0809	-.0208	-.0009	.0001	9.349	.0048	69.0261
694	.806	627.185	-.01	8.24	.7776	.1034	.0243	.0019	.0002	7.520	.0048	87.5402
695	.806	626.899	-.01	9.32	.8028	.1266	.0497	.0045	.0003	6.341	.0048	131.1264
696	.808	628.288	-.01	10.41	.8171	.1492	.0747	.0022	.0002	5.477	.0047	131.8620
697	.810	630.654	-.01	11.51	.8245	.1733	.0911	.0010	.0003	4.758	.0046	114.7586
698	.808	628.823	-.01	12.57	.8456	.1969	.1037	.0011	.0003	4.295	.0046	101.5172
699	.809	629.727	-.00	13.70	.8584	~~.3301~~ .2204	.1135	.0007	.0005	3.895	.0045	101.5172
700	.812	632.117	-.00	15.98	.9106	.2734	.1341	.0007	.0003	3.331	.0043	92.8735
701	.814	634.420	-.01	18.12	.9541	.3259	.1389	.0009	.0004	2.928	.0039	95.6322
702	.807	627.222	-.01	.05	.1146	.0153	-.0526	.0010	.0006	7.467	.0051	9.2689

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 18

POINT	MACH	Q	BETA	ALPHA	CNF	BODY AXIS COEFFICIENTS CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
716	.754	576.339	-.01	.03	.1025	.0164	.0011	.0005	.0012	.0048	0.0000
717	.753	575.816	-.00	1.14	.2046	.0143	.0010	.0006	.0007	.0048	0.0000
718	.753	575.372	-.01	2.28	.3051	.0093	.0011	.0005	.0013	.0049	0.0000
719	.754	576.427	-.01	3.45	.4090	.0017	.0020	.0007	.0012	.0048	0.0000
720	.754	577.049	-.01	4.59	.5063	-.0075	.0018	.0006	.0007	.0048	0.0000
721	.755	577.867	-.00	5.77	.6132	-.0177	.0013	.0004	.0018	.0048	0.0000
722	.755	578.204	-.00	6.95	.7138	-.0220	.0005	.0001	.0017	.0047	0.0000
723	.755	577.799	-.00	8.09	.7542	-.0185	-.0002	.0002	.0012	.0047	0.0000
724	.755	577.584	-.00	9.18	.7909	-.0133	.0027	.0007	.0000	.0046	0.0000
725	.757	579.565	-.01	10.25	.8123	-.0067	.0017	.0006	.0013	.0046	0.0000
726	.754	576.653	-.00	11.35	.8300	-.0008	.0023	.0006	.0001	.0045	0.0000
727	.755	577.484	-.00	12.43	.8492	.0032	.0011	.0005	.0007	.0045	0.0000
728	.756	578.587	-.00	13.58	.8783	.0063	.0007	.0004	.0005	.0045	0.0000
729	.757	580.136	-.00	15.81	.9318	.0095	.0006	.0006	-.0002	.0043	0.0000
730	.759	581.540	-.00	17.93	.9863	.0112	.0010	.0007	.0002	.0041	0.0000
731	.758	580.437	-.00	20.02	1.0487	.0127	.0011	.0007	-.0014	.0035	0.0000
732	.758	580.449	-.00	22.14	1.1199	.0114	.0002	.0009	-.0036	.0027	0.0000
733	.754	577.205	-.00	6.97	.7049	-.0238	.0004	.0001	.0019	.0047	0.0000

TEST= 778 RUN= 18

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
716	.754	576.339	-.01	.03	.1024	.0165	-.0497	.0011	.0005	6.206	.0048	8.6069
717	.753	575.816	-.00	1.14	.2043	.0183	-.0471	.0010	.0006	11.164	.0048	8.4352
718	.753	575.372	-.01	2.28	.3046	.0214	-.0448	.0012	.0005	14.234	.0049	9.2935
719	.754	576.427	-.01	3.45	.4083	.0263	-.0397	.0020	.0005	15.525	.0048	9.5632
720	.754	577.049	-.01	4.59	.5055	.0330	-.0339	.0018	.0005	15.318	.0048	11.1571
721	.755	577.867	-.00	5.77	.6121	.0442	-.0306	.0014	.0002	13.848	.0048	13.4620
722	.755	578.204	-.00	6.95	.7115	.0646	-.0187	.0005	.0000	11.014	.0047	33.9613
723	.755	577.799	-.00	8.09	.7496	.0877	.0178	-.0002	.0003	8.547	.0047	56.2752
724	.755	577.584	-.00	9.18	.7833	.1130	.0496	.0028	.0003	6.932	.0046	74.7884
725	.757	579.565	-.01	10.25	.8009	.1379	.0756	.0018	.0003	5.808	.0046	92.8112
726	.754	576.653	-.00	11.35	.8143	.1626	.0974	.0024	.0002	5.008	.0045	106.4827
727	.755	577.484	-.00	12.43	.8291	.1859	.1123	.0012	.0002	4.460	.0045	93.9770
728	.756	578.587	-.00	13.58	.8528	.2124	.1261	.0008	.0002	4.015	.0045	90.6666
729	.757	580.136	-.00	15.81	.8946	.2631	.1467	.0007	.0004	3.400	.0043	79.8161
730	.759	581.540	-.00	17.93	.9356	.3143	.1539	.0012	.0003	2.977	.0041	82.9425
731	.758	580.437	-.00	20.02	.9815	.3710	.1544	.0012	.0003	2.646	.0035	91.2184
732	.758	580.449	-.00	22.14	1.0335	.4327	.1617	.0005	.0007	2.388	.0027	105.0115
733	.754	577.205	-.00	6.97	.7026	.0618	-.0177	.0004	.0001	11.366	.0047	33.5935

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 19

POINT	MACH	Q	BETA	ALPHA	CNF	BODY AXIS COEFFICIENTS CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
747	.701	523.706	-.00	.05	.0994	.0165	.0012	.0006	-.0002	.0046	0.0000
748	.702	524.803	-.00	1.10	.1914	.0147	.0011	.0006	-.0007	.0046	0.0000
749	.701	523.789	-.00	2.23	.2926	.0097	.0006	.0006	.0005	.0046	0.0000
750	.702	524.098	-.00	3.38	.3877	.0026	.0016	.0006	-.0008	.0046	0.0000
751	.702	524.275	-.00	4.49	.4732	~~.0060~~ -.0060	.0016	.0006	-.0001	.0046	0.0000
752	.702	524.383	-.00	5.61	.5673	-.0164	.0014	.0007	-.0005	.0046	0.0000
753	.702	524.594	-.01	6.78	.6576	-.0206	.0029	.0014	-.0012	.0045	0.0000
754	.702	524.342	-.01	7.90	.7148	-.0173	.0042	.0016	-.0020	.0045	0.0000
755	.704	526.706	-.01	9.03	.7546	-.0116	.0048	.0014	-.0024	.0045	0.0000
756	.703	525.159	-.00	10.08	.7762	-.0040	.0024	.0007	-.0010	.0044	0.0000
757	.703	525.481	-.00	11.22	.8119	-.0013	.0018	.0007	-.0012	.0044	0.0000
758	.704	526.194	-.00	12.31	.8449	.0011	.0015	.0006	-.0016	.0043	0.0000
759	.702	524.225	-.00	13.41	.8645	.0035	.0013	.0005	-.0006	.0043	0.0000
760	.704	526.007	-.00	15.63	.9188	.0073	.0008	.0005	-.0010	.0042	0.0000
761	.705	527.297	-.00	17.77	.9676	.0093	.0007	.0006	-.0014	.0039	0.0000
762	.704	526.648	-.00	19.88	1.0442	.0119	.0009	.0009	-.0019	.0034	0.0000
763	.705	527.451	-.00	21.93	1.1020	.0113	.0003	.0006	-.0031	.0027	0.0000
764	.701	523.053	-.00	-.02	.0948	.0151	.0009	.0006	.0001	.0033	0.0000
765	.699	520.944	-.00	-.02	.0982	.0154	.0018	.0006	-.0007	.0034	0.0000

TEST= 778 RUN= 19

POINT	MACH	Q	BETA	ALPHA	CL	STABILITY AXIS COEFFICIENTS CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
747	.701	523.706	-.00	.05	.0994	.0166	-.0483	.0012	.0006	5.988	.0046	7.7732
748	.702	524.803	-.00	1.10	.1911	.0183	-.0444	.0011	.0006	10.443	.0046	7.2092
749	.701	523.789	-.00	2.23	.2920	.0211	-.0405	.0007	.0005	13.839	.0046	7.4299
750	.702	524.098	-.00	3.38	.3869	.0254	-.0355	.0016	.0005	15.232	.0046	7.3563
751	.702	524.275	-.00	4.49	.4723	.0310	-.0293	.0017	.0005	15.235	.0046	7.8712
752	.702	524.383	-.00	5.61	.5664	.0391	-.0221	.0014	.0006	14.486	.0046	9.0728
753	.702	524.594	-.01	6.78	.6556	.0573	-.0129	.0030	.0011	11.442	.0045	22.9269
754	.702	524.342	-.01	7.90	.7107	.0811	.0161	.0043	.0010	8.763	.0045	62.6506
755	.704	526.706	-.01	9.03	.7474	.1069	.0498	.0049	.0007	6.992	.0044	67.3096
756	.703	525.159	-.00	10.08	.7653	.1320	.0779	.0025	.0003	5.798	.0043	85.0872
757	.703	525.481	-.00	11.22	.7971	.1566	.0983	.0019	.0003	5.090	.0043	90.8496
758	.704	526.194	-.00	12.31	.8257	.1812	.1168	.0016	.0002	4.557	.0042	89.3783
759	.702	524.225	-.00	13.41	.8407	.2039	.1323	.0014	.0002	4.123	.0042	84.3516
760	.704	526.007	-.00	15.63	.8834	.2545	.1537	.0009	.0002	3.471	.0040	78.5892
761	.705	527.297	-.00	17.77	.9193	.3040	.1634	.0008	.0004	3.024	.0038	77.4857
762	.704	526.648	-.00	19.88	.9785	.3662	.1633	.0011	.0005	2.672	.0032	86.1906
763	.705	527.451	-.00	21.93	1.0186	.4221	.1665	.0006	.0005	2.413	.0025	94.1599
764	.701	523.053	-.00	-.02	.0949	.0150	-.0481	.0009	.0006	6.305	.0046	7.2827
765	.699	520.944	-.00	-.02	.0982	.0154	-.0487	.0018	.0006	6.372	.0046	7.2827

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 20

POINT	MACH	Q	BETA	ALPHA	CNF	BODY AXIS COEFFICIENTS CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
779	.495	306.392	-.00	.03	.0848	.0168	.0009	.0007	.0000	.0041	0.0000
780	.498	309.315	-.00	1.09	.1701	.0151	.0012	.0006	-.0008	.0041	0.0000
781	.497	308.785	-.00	2.09	.2522	.0113	.0009	.0005	-.0011	.0041	0.0000
782	.498	309.497	-.00	3.18	.3374	.0053	.0012	.0006	-.0018	.0041	0.0000
783	.498	309.761	-.00	4.17	.4114	-.0020	.0012	.0004	.0000	.0041	0.0000
784	.498	309.856	-.00	5.28	.4894	-.0120	.0014	.0006	-.0007	.0041	0.0000
785	.498	309.159	-.00	6.37	.5723	-.0230	.0014	.0009	-.0013	.0041	0.0000
786	.497	308.760	-.00	7.45	.6472	-.0292	.0018	.0022	-.0046	.0040	0.0000
787	.497	308.867	-.00	8.55	.7108	-.0235	.0030	.0014	-.0042	.0040	0.0000
788	.497	308.355	-.00	9.61	.7517	-.0171	.0028	.0011	-.0028	.0039	0.0000
789	.497	308.992	-.00	10.72	.7861	-.0125	.0022	.0007	-.0032	.0038	0.0000
790	.497	308.562	-.00	11.80	.8198	-.0094	.0019	.0004	-.0014	.0038	0.0000
791	.500	311.689	.00	12.91	.8543	-.0064	.0011	.0002	-.0015	.0038	0.0000
792	.498	309.435	.00	15.13	.9210	-.0015	.0008	.0004	-.0024	.0036	0.0000
793	.499	310.101	-.00	17.22	.9761	.0018	.0005	.0003	-.0006	.0034	0.0000
794	.499	310.864	-.00	19.27	1.0269	.0043	.0003	.0003	-.0010	.0032	0.0000
795	.498	309.058	-.00	21.28	1.0786	.0072	-.0001	.0006	-.0007	.0026	0.0000
796	.497	308.779	-.00	-.02	.0880	.0173	.0009	.0005	.0000	.0041	0.0000

TEST= 778 RUN= 20

POINT	MACH	Q	BETA	ALPHA	CL	STABILITY AXIS COEFFICIENTS CD	CPM	CLS	CNS	L/D	STAB.PRESS.COEFF CDB	PB-1
779	.495	306.392	-.00	.03	.0848	.0168	-.0434	.0009	.0007	5.048	.0041	4.4994
780	.498	309.315	-.00	1.09	.1697	.0183	-.0390	.0012	.0006	9.273	.0041	4.5485
781	.497	308.785	-.00	2.09	.2517	.0204	-.0341	.0009	.0005	12.338	.0041	4.5117
782	.498	309.497	-.00	3.18	.3367	.0240	-.0287	.0012	.0006	14.029	.0041	4.3400
783	.498	309.761	-.00	4.17	.4106	.0278	-.0233	.0012	.0003	14.770	.0041	4.0703
784	.498	309.856	-.00	5.28	.4886	.0330	-.0175	.0014	.0005	14.806	.0040	4.2665
785	.498	309.159	-.00	6.37	.5715	.0406	-.0100	.0015	.0007	14.076	.0040	4.3891
786	.497	308.760	-.00	7.45	.6457	.0549	-.0020	.0021	.0020	11.761	.0040	10.3724
787	.497	308.867	-.00	8.55	.7066	.0824	.0223	.0031	.0009	8.575	.0039	39.7850
788	.497	308.355	-.00	9.61	.7443	.1087	.0547	.0029	.0006	6.847	.0039	56.3978
789	.497	308.992	-.00	10.72	.7750	.1340	.0851	.0023	.0003	5.784	.0038	67.4322
790	.497	308.562	-.00	11.80	.8047	.1584	.1087	.0019	-.0000	5.080	.0037	68.4130
791	.500	311.689	.00	12.91	.8345	.1846	.1281	.0011	-.0000	4.521	.0037	74.0528
792	.498	309.435	.00	15.13	.8899	.2389	.1593	.0009	.0002	3.725	.0035	75.4015
793	.499	310.101	-.00	17.22	.9323	.2907	.1795	.0006	.0001	3.207	.0033	69.7617
794	.499	310.864	-.00	19.27	.9686	.3430	.1928	.0004	.0002	2.824	.0030	66.9418
795	.498	309.058	-.00	21.28	1.0029	.3981	.1883	.0001	.0006	2.519	.0024	71.1103
796	.497	308.779	-.00	-.02	.0880	.0173	-.0439	.0009	.0005	5.087	.0041	4.5975

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 21

POINT	MACH	Q	BETA	ALPHA	CNF	BODY AXIS COEFFICIENTS CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
810	.292	118.610	-.00	-.00	.0603	.0165	.0006	.0006	.0004	.0038	0.0000
811	.293	119.382	-.00	1.01	.1376	.0147	.0012	.0004	-.0007	.0038	0.0000
812	.292	119.286	-.00	2.00	.2151	.0116	.0014	.0006	-.0012	.0038	0.0000
813	.293	119.866	-.00	3.02	.3008	.0065	.0015	.0004	-.0019	.0038	0.0000
814	.294	120.543	-.00	4.03	.3880	.0006	.0019	.0006	-.0027	.0038	0.0000
815	.293	119.869	.00	5.06	.4351	-.0090	.0015	.0004	-.0027	.0039	0.0000
816	.293	119.678	.00	6.11	.5125	-.0186	.0014	.0006	-.0033	.0039	0.0000
817	.293	119.778	-.00	7.12	.5789	-.0304	.0014	.0012	-.0038	.0037	0.0000
818	.293	119.791	-.00	8.19	.6544	-.0373	.0002	.0027	-.0094	.0037	0.0000
819	.292	119.147	-.00	9.26	.7190	-.0343	.0057	.0040	-.0149	.0037	0.0000
820	.292	119.080	-.00	10.31	.7587	-.0317	.0086	.0041	-.0123	.0036	0.0000
821	.291	118.437	-.00	11.42	.8013	-.0294	.0086	.0034	-.0132	.0036	0.0000
822	.291	118.476	.00	12.54	.8380	-.0240	.0068	.0018	-.0125	.0035	0.0000
823	.291	118.464	.00	14.67	.9074	-.0108	.0010	.0001	-.0090	.0034	0.0000
824	.292	119.116	.00	16.72	.9563	-.0058	.0008	-.0001	-.0097	.0031	0.0000
825	.293	119.781	.00	18.80	1.0320	-.0022	.0007	.0002	-.0048	.0031	0.0000
826	.292	119.074	.00	20.83	1.0603	-.0010	.0007	.0003	-.0050	.0026	0.0000
827	.292	118.727	.00	8.21	.6416	-.0387	.0002	.0026	-.0153	.0037	0.0000

TEST= 778 RUN= 21

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
810	.292	118.610	-.00	-.00	.0603	.0165	-.0460	.0006	.0006	3.655	.0038	1.8267
811	.293	119.382	-.00	1.01	.1373	.0171	-.0376	.0012	.0003	8.029	.0038	1.8145
812	.292	119.286	-.00	2.00	.2145	.0191	-.0365	.0014	.0005	11.230	.0038	2.1823
813	.293	119.866	-.00	3.02	.3000	.0223	-.0306	.0016	.0003	13.457	.0038	1.8022
814	.294	120.543	-.00	4.03	.3652	.0263	-.0235	.0020	.0004	13.886	.0038	2.1455
815	.293	119.869	.00	5.06	.4342	.0294	-.0202	.0015	.0002	14.772	.0038	1.6919
816	.293	119.678	.00	6.11	.5117	.0361	-.0122	.0015	.0005	14.175	.0038	1.6061
817	.293	119.778	-.00	7.12	.5782	.0415	-.0053	.0015	.0011	13.923	.0037	2.2558
818	.293	119.791	-.00	8.19	.6532	.0563	.0026	.0006	.0026	11.602	.0037	7.4909
819	.292	119.147	-.00	9.26	.7154	.0817	.0157	.0063	.0030	8.756	.0037	20.4749
820	.292	119.080	-.00	10.31	.7524	.1046	.0438	.0092	.0025	7.193	.0035	28.2602
821	.291	118.437	-.00	11.42	.7915	.1299	.0686	.0091	.0016	6.093	.0035	34.2678
822	.291	118.476	.00	12.54	.8236	.1584	.0965	.0071	.0003	5.199	.0034	40.0302
823	.291	118.464	.00	14.67	.8809	.2193	.1450	.0010	-.0001	4.017	.0032	39.8463
824	.292	119.116	.00	16.72	.9178	.2697	.1699	.0007	-.0004	3.403	.0030	36.9651
825	.293	119.781	.00	18.80	.9780	.3304	.1977	.0008	-.0000	2.960	.0028	33.5935
826	.292	119.074	.00	20.83	.9917	.3761	.2048	.0008	.0000	2.637	.0036	33.7774
827	.292	118.727	.00	8.21	.6405	.0533	-.0030	.0006	.0025	12.023	.0037	7.7115

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 22

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
848	.807	626.795	-.00	.21	.1248	.0140	.0018	.0006	.0006	.0051	0.0000
849	.807	627.301	-.01	1.36	.2442	.0116	.0018	.0006	.0012	.0051	0.0000
850	.806	625.813	-.00	2.56	.3556	.0060	.0020	.0006	.0004	.0050	0.0000
851	.808	627.974	-.01	3.69	.4612	-.0021	.0026	.0006	.0005	.0050	0.0000
852	.804	624.611	-.00	4.81	.5575	-.0103	.0020	.0002	.0004	.0050	0.0000
853	.808	627.813	-.00	6.00	.6802	-.0146	-.0002	-.0002	.0016	.0049	0.0000
855	.806	626.056	-.01	8.26	.7837	-.0101	.0044	.0008	-.0001	.0048	0.0000
854	.808	627.635	-.00	7.15	.7466	-.0132	-.0009	.0001	.0011	.0049	0.0000
856	.807	627.383	-.01	9.35	.8208	-.0054	.0062	.0008	.0009	.0048	0.0000
857	.809	628.807	-.00	10.44	.8342	.0010	.0029	.0004	.0008	.0048	0.0000
858	.808	628.316	-.00	11.53	.8418	.0044	.0009	.0003	.0009	.0047	0.0000
859	.810	629.714	-.00	12.58	.8614	.0080	.0012	.0004	.0002	.0047	0.0000
860	.810	629.468	-.00	13.73	.8882	.0102	.0009	.0004	.0000	.0047	0.0000
861	.809	628.828	-.00	14.81	.9119	.0085	.0006	.0005	-.0001	.0046	0.0000
862	.810	630.143	-.00	15.97	.9477	.0113	.0007	.0006	-.0006	.0045	0.0000
863	.811	630.835	-.00	18.13	1.0189	.0121	.0006	.0006	-.0011	.0041	0.0000
864	.812	631.143	.00	20.23	1.0832	.0124	.0005	.0006	-.0035	.0035	0.0000
865	.806	625.828	-.00	.07	.1289	.0122	.0021	.0006	.0004	.0050	0.0000
866	.806	625.904	-.01	.07	.1303	.0121	.0016	.0006	.0008	.0050	0.0000

TEST= 778 RUN= 22

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
848	.807	626.795	-.00	.21	.1248	.0145	-.0587	.0018	.0005	8.607	.0051	9.3180
849	.807	627.301	-.01	1.36	.2439	.0174	-.0579	.0018	.0006	14.017	.0050	10.3233
850	.806	625.813	-.00	2.56	.3550	.0219	-.0550	.0021	.0005	16.210	.0050	11.7210
851	.808	627.974	-.01	3.69	.4605	.0276	-.0508	.0026	.0004	16.685	.0050	16.4045
852	.804	624.611	-.00	4.81	.5566	.0366	-.0446	.0020	.0001	15.208	.0049	16.5517
853	.808	627.813	-.00	6.00	.6783	.0566	-.0430	-.0002	-.0002	16.984	.0049	26.4825
855	.806	626.056	-.01	8.26	.7774	.1025	.0202	.0045	.0001	7.584	.0048	176.3060
854	.808	627.635	-.00	7.15	.7428	.0798	-.0133	-.0008	.0002	9.308	.0048	60.9342
856	.807	627.383	-.01	9.35	.8113	.1280	.0421	.0062	-.0002	6.338	.0048	249.7471
857	.809	628.807	-.00	10.44	.8207	.1520	.0680	.0029	-.0002	5.399	.0047	218.1149
858	.808	628.316	-.00	11.53	.8245	.1725	.0878	.0010	.0001	4.780	.0046	130.9425
859	.810	629.714	-.00	12.58	.8395	.1955	.1010	.0013	.0001	4.294	.0046	105.1954
860	.810	629.468	-.00	13.73	.8610	.2206	.1124	.0010	.0002	3.903	.0045	101.8850
861	.809	628.828	-.00	14.81	.8794	.1235	.1235	.0007	.0004	3.645	.0046	95.2643
862	.810	630.143	-.00	15.97	.9087	.2716	.1326	.0008	.0004	3.346	.0043	90.1149
863	.811	630.835	-.00	18.13	.9651	.3285	.1403	.0008	.0004	2.938	.0039	93.0574
864	.812	631.143	.00	20.23	1.0121	.3861	.1367	.0007	.0004	2.621	.0035	107.9540
865	.806	625.828	-.00	.07	.1289	.0124	-.0591	.0021	.0006	10.416	.0050	9.4406
866	.806	625.904	-.01	.07	.1303	.0123	-.0582	.0016	.0006	10.587	.0050	9.4406

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 23

POINT	MACH	Q	BETA	ALPHA	CNF	BODY AXIS COEFFICIENTS CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
883	.753	575.300	-.00	-.02	.1176	.0143	.0018	.0005	-.0006	.0048	0.0000
884	.754	575.821	-.00	1.13	.2255	.0118	.0017	.0006	-.0010	.0048	0.0000
885	.754	575.825	-.00	2.30	.3253	.0067	.0014	.0006	-.0001	.0049	0.0000
886	.753	574.946	-.00	3.43	.4165	-.0004	.0023	.0006	-.0004	.0048	0.0000
887	.755	576.701	-.00	4.58	.5169	-.0093	.0019	.0005	-.0006	.0048	0.0000
888	.753	574.748	-.00	5.76	.6088	-.0187	.0016	.0003	.0003	.0048	0.0000
889	.753	574.932	-.00	6.94	.7097	-.0224	.0003	-.0001	.0007	.0047	0.0000
890	.755	576.794	-.00	8.06	.7606	-.0184	.0013	.0004	-.0002	.0047	0.0000
891	.755	577.030	-.00	9.19	.7967	-.0131	.0040	.0008	-.0011	.0046	0.0000
892	.755	576.906	-.00	10.25	.8190	-.0068	.0025	.0004	-.0006	.0046	0.0000
893	.756	578.128	-.00	11.35	.8296	0	.0017	.0005	-.0005	.0045	0.0000
894	.755	577.625	-.00	12.44	.8571	.0034	.0011	.0004	-.0005	.0045	0.0000
895	.756	578.562	-.00	13.58	.8809	.0066	.0012	.0003	-.0011	.0045	0.0000
896	.756	578.499	-.00	15.82	.9288	.0088	.0007	.0005	-.0014	.0043	0.0000
897	.757	579.504	-.00	17.97	.9999	.0106	.0008	.0003	-.0008	.0041	0.0000
898	.759	581.210	.00	20.06	1.0555	.0127	.0010	.0005	-.0025	.0035	0.0000
899	.759	580.843	.00	22.15	1.1273	.0108	-.0000	.0007	-.0058	.0027	0.0000
900	.755	577.234	-.00	.04	.1190	.0128	.0016	.0005	-.0004	.0048	0.0000

TEST= 778 RUN= 23

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
883	.753	575.300	-.00	-.02	.1176	.0143	-.0552	.0018	.0005	8.224	.0048	7.6015
884	.754	575.821	-.00	1.13	.2253	.0163	-.0525	.0017	.0006	13.822	.0048	8.1655
885	.754	575.825	-.00	2.30	.3249	.0198	-.0493	.0014	.0005	16.409	.0049	8.6559
886	.753	574.946	-.00	3.43	.4159	.0246	-.0386	.0023	.0004	16.907	.0048	9.8574
887	.755	576.701	-.00	4.58	.5161	.0320	-.0336	.0019	.0004	16.128	.0048	11.6720
888	.753	574.748	-.00	5.76	.6078	.0425	-.0280	.0017	.0001	14.301	.0047	13.7808
889	.753	574.932	-.00	6.94	.7076	.0635	-.0154	.0003	-.0001	11.143	.0047	34.3904
890	.755	576.794	-.00	8.06	.7561	.0884	.0184	.0014	.0002	8.553	.0046	56.8883
891	.755	577.030	-.00	9.19	.7891	.1142	.0481	.0040	.0001	6.910	.0046	79.0796
892	.755	576.906	-.00	10.25	.8077	.1390	.0727	.0026	-.0001	5.811	.0045	96.1215
893	.756	578.128	-.00	11.35	.8139	.1632	.0967	.0017	.0001	4.987	.0044	92.9338
894	.755	577.625	-.00	12.44	.8368	.1879	.1125	.0011	.0001	4.453	.0044	89.3783
895	.756	578.562	-.00	13.58	.8553	.2133	.1257	.0013	-.0000	4.010	.0043	93.0574
896	.756	578.499	-.00	15.82	.8918	.2618	.1451	.0008	.0003	3.406	.0041	82.0230
897	.757	579.504	-.00	17.97	.9484	.3187	.1541	.0008	.0001	2.976	.0039	81.8391
898	.759	581.210	.00	20.06	.9877	.3739	.1514	.0011	.0001	2.642	.0033	99.3103
899	.759	580.843	.00	22.15	1.0404	.4350	.1606	.0002	.0006	2.393	.0025	107.7701
900	.755	577.234	-.00	.04	.1190	.0129	-.0556	.0016	.0005	9.252	.0048	8.0919

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 24

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
914	.702	524.324	-.00	.03	.1152	.0144	.0017	.0007	-.0005	.0046	0.0000
915	.702	523.938	-.00	1.10	.2117	.0122	.0011	.0006	-.0006	.0046	0.0000
916	.701	522.929	-.00	2.25	.3095	.0075	.0018	.0007	-.0004	.0046	0.0000
917	.701	522.938	-.00	3.36	.3980	.0008	.0026	.0006	-.0004	.0046	0.0000
918	.702	523.667	-.00	4.47	.4804	-.0074	.0018	.0006	-.0003	.0046	0.0000
919	.701	523.152	-.00	5.61	.5732	-.0175	.0016	.0007	-.0008	.0046	0.0000
920	.701	523.444	-.00	6.79	.6634	-.0218	.0024	.0011	-.0021	.0045	0.0000
921	.702	523.734	-.00	7.91	.7276	-.0196	.0015	.0007	-.0009	.0045	0.0000
922	.701	523.380	-.00	9.01	.7540	-.0105	-.0006	.0002	-.0001	.0044	0.0000
923	.703	525.279	-.00	10.12	.7863	-.0052	.0041	.0007	-.0011	.0044	0.0000
924	.702	524.507	-.00	11.22	.8186	-.0014	.0026	.0005	-.0019	.0043	0.0000
925	.703	524.746	-.00	12.31	.8474	.0011	.0016	.0004	-.0003	.0043	0.0000
926	.703	525.085	-.00	13.46	.8785	.0031	.0012	.0004	-.0005	.0043	0.0000
927	.704	526.313	-.00	15.67	.9292	.0068	.0008	.0005	-.0010	.0042	0.0000
928	.705	527.522	-.00	17.78	.9783	.0089	.0008	.0006	-.0015	.0039	0.0000
929	.706	528.010	-.00	19.85	1.0333	.0118	.0012	.0006	-.0020	.0034	0.0000
930	.707	529.005	-.00	21.96	1.0989	.0108	.0004	.0006	-.0031	.0027	0.0000
931	.702	523.693	-.01	6.82	.6629	-.0238	.0023	.0011	-.0007	.0045	0.0000

TEST= 778 RUN= 24

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	L/D	STAB.PRESS.COEFF CDB	PB-1
914	.702	524.324	-.00	.03	.1152	.0145	-.0535	.0017	.0007	7.945	.0046	7.5154
915	.702	523.938	-.00	1.10	.2115	.0163	-.0483	.0011	.0006	12.975	.0046	7.2579
916	.701	522.929	-.00	2.25	.3090	.0196	-.0447	.0018	.0006	15.765	.0046	7.2334
917	.701	522.938	-.00	3.36	.3974	.0241	-.0349	.0026	.0005	16.490	.0046	7.7851
918	.702	523.667	-.00	4.47	.4796	.0300	-.0272	.0018	.0005	15.987	.0046	8.4594
919	.701	523.152	-.00	5.61	.5724	.0386	-.0209	.0017	.0006	14.829	.0046	8.6923
920	.701	523.444	-.00	6.79	.6616	.0568	-.0124	.0025	.0008	11.648	.0045	21.5170
921	.702	523.734	-.00	7.91	.7237	.0808	.0136	.0016	.0005	8.957	.0045	61.6698
922	.701	523.380	-.00	9.01	.7467	.1078	.0487	-.0006	.0003	6.927	.0044	82.3908
923	.703	525.279	-.00	10.12	.7755	.1330	.0727	.0042	-.0000	5.831	.0043	98.2069
924	.702	524.507	-.00	11.22	.8038	.1578	.0967	.0027	.0000	5.094	.0043	102.4368
925	.703	524.746	-.00	12.31	.8282	.1817	.1155	.0016	.0001	4.558	.0042	86.9885
926	.703	525.085	-.00	13.46	.8542	.2074	.1314	.0013	.0001	4.119	.0042	84.5977
927	.704	526.313	-.00	15.67	.8935	.2575	.1525	.0009	.0003	3.470	.0040	80.0000
928	.705	527.522	-.00	17.78	.9295	.3073	.1630	.0009	.0003	3.025	.0038	77.4253
929	.706	528.010	-.00	19.85	.9685	.3621	.1586	.0013	.0002	2.675	.0032	93.2414
930	.707	529.005	-.00	21.96	1.0157	.4208	.1640	.0006	.0004	2.414	.0025	95.0804
931	.702	523.693	-.01	6.82	.6611	.0550	-.0125	.0024	.0008	12.018	.0045	21.5170

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 25

POINT	MACH	Q	BETA	ALPHA	CNF	BODY AXIS COEFFICIENTS CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
945	.498	309.310	-.00	.01	.0998	.0147	.0018	.0007	-.0007	.0041	0.0000
946	.498	309.930	-.00	1.07	.1854	.0128	.0022	.0006	-.0016	.0041	0.0000
947	.498	309.930	-.00	2.07	.2630	.0089	.0015	.0006	-.0015	.0041	0.0000
948	.498	310.113	-.00	3.15	.3411	.0032	.0021	.0006	-.0026	.0041	0.0000
949	.498	309.941	.00	4.17	.4148	-.0035	.0010	.0004	-.0022	.0041	0.0000
950	.499	310.656	.00	5.26	.4954	-.0132	.0009	.0004	-.0027	.0041	0.0000
951	.498	310.046	.00	6.34	.5749	-.0244	.0019	.0008	-.0041	.0041	0.0000
952	.498	310.088	-.00	7.42	.6516	-.0309	.0013	.0019	-.0042	.0040	0.0000
953	.498	309.657	.00	8.53	.7195	-.0273	.0009	.0005	-.0046	.0040	0.0000
954	.497	309.051	.00	9.59	.7730	-.0229	-.0042	-.0010	-.0010	.0039	0.0000
955	.499	310.415	-.00	10.70	.7937	-.0139	.0022	.0007	-.0033	.0038	0.0000
956	.498	309.807	.00	11.75	.8276	-.0106	.0016	.0004	-.0036	.0038	0.0000
957	.497	308.765	-.00	12.87	.8623	-.0078	.0011	.0004	-.0014	.0038	0.0000
958	.497	309.086	.00	15.10	.9259	-.0024	.0008	.0003	-.0023	.0036	0.0000
959	.500	312.248	.00	17.20	.9842	.0012	.0004	.0003	-.0029	.0034	0.0000
960	.499	310.524	.00	19.25	1.0359	.0040	.0007	.0005	-.0037	.0032	0.0000
961	.499	310.675	.00	21.26	1.0848	.0064	.0006	.0006	-.0037	.0026	0.0000
962	.497	308.603	-.00	-.04	.1001	.0154	.0018	.0005	-.0007	.0041	0.0000

TEST= 778 RUN= 25

POINT	MACH	Q	BETA	ALPHA	CL	STABILITY AXIS COEFFICIENTS CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
945	.498	309.310	-.00	.01	.0998	.0147	-.0478	.0018	.0007	6.789	.0041	4.3891
946	.498	309.930	-.00	1.07	.1852	.0163	-.0421	.0022	.0006	11.362	.0041	4.6098
947	.498	309.930	-.00	2.07	.2626	.0184	-.0362	.0015	.0005	14.272	.0041	4.3768
948	.498	310.113	-.00	3.15	.3404	.0220	-.0261	.0021	.0005	15.473	.0041	4.7201
949	.498	309.941	.00	4.17	.4141	.0266	-.0193	.0010	.0003	15.568	.0041	4.6711
950	.499	310.656	.00	5.26	.4947	.0321	-.0141	.0010	.0004	15.411	.0040	4.6220
951	.498	310.046	.00	6.34	.5743	.0392	-.0082	.0020	.0006	14.651	.0040	4.3033
952	.498	310.088	-.00	7.42	.6503	.0536	.0002	.0015	.0017	12.132	.0040	9.7594
953	.498	309.657	.00	8.53	.7159	.0796	.0181	.0009	.0003	8.994	.0039	31.5705
954	.497	309.051	.00	9.59	.7663	.1063	.0424	-.0043	-.0003	7.209	.0039	48.3060
955	.499	310.415	-.00	10.70	.7828	.1336	.0860	.0023	.0003	5.859	.0038	66.3288
956	.498	309.807	.00	11.75	.8129	.1581	.1095	.0017	.0000	5.142	.0037	67.0644
957	.497	308.765	-.00	12.87	.8429	.1845	.1279	.0012	.0001	4.569	.0037	70.8651
958	.497	309.086	.00	15.10	.8951	.2388	.1599	.0009	.0000	3.748	.0035	73.6850
959	.500	312.248	.00	17.20	.9404	.2922	.1813	.0005	.0002	3.218	.0033	68.6582
960	.499	310.524	.00	19.25	.9772	.3453	.1937	.0008	.0002	2.830	.0030	68.2904
961	.499	310.675	.00	21.26	1.0092	.3993	.1893	.0008	.0003	2.527	.0024	71.2329
962	.497	308.603	-.00	-.04	.1001	.0154	-.0465	.0018	.0005	6.500	.0041	4.5117

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 26

POINT	MACH	Q	BETA	ALPHA	CNF	BODY AXIS COEFFICIENTS CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
978	.292	119.381	-.00	.04	.0802	.0146	.0015	.0002	-.0004	.0038	0.0000
979	.293	119.478	-.00	1.02	.1476	.0134	.0016	.0004	-.0009	.0038	0.0000
980	.293	119.478	-.00	2.01	.2435	.0096	.0014	.0007	-.0014	.0038	0.0000
981	.293	119.479	-.00	3.02	.3113	.0038	.0015	.0004	-.0020	.0038	0.0000
982	.293	120.157	-.00	4.03	.3763	-.0014	.0015	.0006	-.0025	.0038	0.0000
983	.294	120.160	-.00	5.04	.4525	-.0097	.0014	.0006	.0030	.0039	0.0000
984	.294	120.162	-.00	6.11	.5199	-.0198	.0021	.0007	.0020	.0039	0.0000
985	.294	120.166	-.00	7.12	.5866	-.0311	.0014	.0013	-.0038	.0037	0.0000
986	.293	119.599	-.00	8.19	.6560	-.0385	.0013	.0023	-.0043	.0037	0.0000
987	.292	118.853	-.00	9.26	.7289	-.0390	.0033	.0029	-.0067	.0037	0.0000
988	.292	118.894	-.00	10.39	.7986	-.0334	.0086	.0038	-.0126	.0036	0.0000
989	.291	118.342	-.00	11.46	.8120	-.0314	.0091	.0034	-.0135	.0036	0.0000
990	.291	118.287	-.00	12.52	.8679	-.0258	.0068	.0020	-.0067	.0035	0.0000
991	.292	119.047	-.00	14.73	.9118	-.0123	.0015	.0004	.0027	.0034	0.0000
992	.293	119.799	-.00	16.79	.9687	-.0071	.0008	.0002	.0022	.0031	0.0000
993	.293	119.882	-.00	18.80	1.0405	-.0029	.0011	.0004	.0008	.0030	0.0000
994	.293	119.959	-.00	20.84	1.0804	.0004	.0006	.0005	.0008	.0026	0.0000
995	.292	118.923	-.00	8.21	.6496	-.0393	.0009	.0023	-.0040	.0037	0.0000

TEST= 778 RUN= 26

POINT	MACH	Q	BETA	ALPHA	CL	STABILITY AXIS COEFFICIENTS CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
978	.292	119.381	-.00	.04	.0802	.0146	-.0465	.0015	.0002	5.493	.0038	1.7409
979	.293	119.478	-.00	1.02	.1474	.0160	-.0425	.0017	.0003	9.213	.0038	1.8390
980	.293	119.478	-.00	2.01	.2430	.0181	-.0318	.0014	.0006	13.425	.0038	1.9493
981	.293	119.479	-.00	3.02	.3106	.0202	-.0277	.0016	.0003	15.346	.0038	1.8880
982	.293	120.157	-.00	4.03	.3755	.0251	-.0201	.0015	.0005	14.960	.0038	1.8758
983	.294	120.160	-.00	5.04	.4516	.0301	-.0151	.0015	.0004	15.019	.0038	1.6919
984	.294	120.162	-.00	6.11	.5191	.0356	-.0077	.0021	.0005	14.581	.0038	1.6919
985	.294	120.166	-.00	7.12	.5859	.0418	-.0042	.0015	.0011	14.022	.0037	1.7777
986	.293	119.599	-.00	8.19	.6550	.0553	.0026	.0017	.0021	11.844	.0037	7.3315
987	.292	118.853	-.00	9.26	.7259	.0787	.0132	.0037	.0024	9.224	.0037	15.3011
988	.292	118.894	-.00	10.39	.7918	.1112	.0516	.0092	.0022	7.121	.0035	29.4250
989	.291	118.342	-.00	11.46	.8024	.1305	.0698	.0096	.0015	6.149	.0035	33.4709
990	.291	118.287	-.00	12.52	.8532	.1629	.1034	.0071	.0005	5.238	.0034	36.7199
991	.292	119.047	-.00	14.73	.8854	.2199	.1452	.0015	-.0000	4.026	.0032	41.0723
992	.293	119.799	-.00	16.79	.9298	.2731	.1743	.0008	-.0001	3.405	.0030	37.6394
993	.293	119.882	-.00	18.80	.9864	.3325	.1983	.0012	-.0000	2.967	.0028	33.8387
994	.293	119.959	-.00	20.84	1.0100	.3848	.2085	.0007	.0003	2.625	.0024	32.4901
995	.292	118.923	-.00	8.21	.6485	.0539	.0050	.0012	.0022	12.025	.0037	7.2579

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 27

POINT	MACH	Q	BETA	ALPHA	CNF	BODY AXIS COEFFICIENTS CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
10	.932	731.969	-.01	.15	.1702	.0213	.0020	.0006	.0010	.0064	0.0000
11	.931	731.520	-.01	1.38	.2881	.0269	.0011	.0005	.0013	.0064	0.0000
12	.931	731.624	-.01	2.51	.3794	.0243	.0011	.0006	.0008	.0063	0.0000
13	.933	732.671	-.01	3.72	.4848	.0230	.0016	.0005	.0007	.0063	0.0000
14	.933	733.224	-.01	4.91	.5751	.0219	.0009	.0005	.0009	.0063	0.0000
15	.934	733.865	-.01	6.08	.6583	.0202	.0016	.0003	.0018	.0062	0.0000
16	.934	733.924	-.01	7.30	.7470	.0183	.0030	.0004	.0019	.0062	0.0000
17	.935	734.619	-.01	8.49	.8212	.0169	.0051	.0007	.0005	.0062	0.0000
18	.939	737.405	-.01	9.68	.9039	.0163	.0070	.0007	.0014	.0060	0.0000
19	.936	734.948	-.01	10.85	.9645	.0146	.0120	.0010	.0011	.0058	0.0000

TEST= 778 RUN= 27

POINT	MACH	Q	BETA	ALPHA	CL	STABILITY AXIS COEFFICIENTS CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
10	.932	731.969	-.01	.15	.1702	.0217	-.0924	.0020	.0006	7.843	.0064	16.3676
11	.931	731.520	-.01	1.38	.2875	.0309	-.0946	.0011	.0005	9.304	.0064	29.7315
12	.931	731.624	-.01	2.51	.3781	.0410	-.0913	.0011	.0005	9.222	.0063	37.4555
13	.933	732.671	-.01	3.72	.4824	.0544	-.0895	.0017	.0004	8.868	.0063	54.8040
14	.933	733.224	-.01	4.91	.5713	.0709	-.0884	.0009	.0004	8.058	.0063	60.8116
15	.934	733.865	-.01	6.08	.6527	.0899	-.0841	.0016	.0002	7.260	.0062	91.9537
16	.934	733.924	-.01	7.30	.7390	.1130	-.0783	.0031	.0001	6.540	.0061	114.5131
17	.935	734.619	-.01	8.49	.8101	.1379	-.0705	.0051	-.0000	5.875	.0061	120.6433
18	.939	737.405	-.01	9.68	.8888	.1680	-.0721	.0070	-.0005	5.290	.0059	155.4631
19	.936	734.948	-.01	10.85	.9450	.1960	-.0537	.0120	-.0013	4.821	.0057	149.5781

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 28

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
34	.861	675.312	-.01	-.05	.1452	.0155	.0017	.0006	.0007	.0055	0.0000
35	.863	676.704	-.01	1.15	.2696	.0134	.0020	.0007	.0020	.0054	0.0000
36	.863	676.745	-.01	2.33	.3975	.0093	.0016	.0006	.0019	.0054	0.0000
37	.864	677.383	-.01	3.56	.5288	.0059	.0015	.0003	.0016	.0053	0.0000
38	.864	677.802	-.00	4.72	.6267	.0039	.0029	.0004	.0011	.0053	0.0000
39	.863	676.909	-.01	5.88	.7051	.0024	.0028	.0008	.0004	.0053	0.0000
40	.865	678.505	-.01	7.04	.7720	.0005	.0028	.0013	-.0006	.0053	0.0000
41	.864	677.645	-.01	8.20	.8207	.0006	.0048	.0017	-.0012	.0052	0.0000
42	.866	679.285	-.01	9.31	.8615	.0016	.0066	.0017	-.0017	.0051	0.0000
43	.863	676.828	-.02	10.38	.8685	.0023	.0118	.0022	-.0008	.0051	0.0000
44	.861	675.065	-.01	-.09	.1418	.0139	.0017	.0007	.0007	.0055	0.0000

TEST= 778 RUN= 28

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
34	.861	675.312	-.01	-.05	.1453	.0154	-.0653	.0017	.0006	9.435	.0055	11.9417
35	.863	676.704	-.01	1.15	.2693	.0188	-.0672	.0020	.0007	14.324	.0054	15.9877
36	.863	676.745	-.01	2.33	.3968	.0254	-.0707	.0016	.0005	15.622	.0054	17.5325
37	.864	677.383	-.01	3.56	.5276	.0387	-.0798	.0016	.0002	13.633	.0053	19.8618
38	.864	677.802	-.00	4.72	.6244	.0554	-.0823	.0029	.0001	11.271	.0053	31.3253
39	.863	676.909	-.01	5.88	.7014	.0746	-.0683	.0029	.0005	9.402	.0053	60.0760
40	.865	678.505	-.01	7.04	.7665	.0951	-.0453	.0029	.0009	8.060	.0052	82.7577
41	.864	677.645	-.01	8.20	.8127	.1175	-.0176	.0050	.0010	6.917	.0052	82.3899
42	.866	679.285	-.01	9.31	.8504	.1408	.0072	.0068	.0006	6.040	.0050	84.7194
43	.863	676.828	-.02	10.38	.8544	.1587	.0344	.0120	.0000	5.384	.0050	135.6011
44	.861	675.065	-.01	-.09	.1419	.0137	-.0657	.0017	.0007	10.356	.0055	11.9662

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 29

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
109	.930	729.125	-.01	-.14	-.0020	.0199	.0003	.0006	.0005	.0062	0.0000
110	.931	729.723	-.01	1.07	.1246	.0203	-.0001	.0005	.0005	.0062	0.0000
111	.930	729.380	-.01	2.29	.2342	.0199	.0003	.0005	.0007	.0061	0.0000
112	.931	729.563	-.01	3.50	.3362	.0186	-.0005	.0004	.0009	.0062	0.0000
113	.933	731.256	-.01	4.69	.4541	.0185	-.0008	.0005	.0009	.0062	0.0000
114	.935	732.665	-.01	5.93	.5647	.0202	.0006	.0005	.0013	.0060	0.0000
115	.935	732.667	-.01	7.15	.6633	.0216	.0001	.0005	.0013	.0060	0.0000
116	.935	732.727	-.00	8.31	.7265	.0224	-.0002	.0003	.0007	.0060	0.0000
117	.936	733.592	-.00	9.54	.8057	.0231	.0002	.0002	.0007	.0058	0.0000
118	.937	734.470	-.00	10.69	.8752	.0234	.0007	.0002	.0008	.0057	0.0000
119	.931	729.741	-.01	-.05	-.0024	.0200	-.0002	.0005	.0010	.0061	0.0000

TEST= 778 RUN= 29

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
109	.930	729.125	-.01	-.14	-.0020	.0199	.0065	.0003	.0006	-.099	.0062	11.5535
110	.931	729.723	-.01	1.07	.1242	.0227	-.0016	-.0001	.0005	5.482	.0062	15.1970
111	.930	729.380	-.01	2.29	.2332	.0292	-.0048	.0003	.0005	7.992	.0061	21.8726
112	.931	729.563	-.01	3.50	.3345	.0390	.0002	-.0005	.0005	8.568	.0062	20.8539
113	.933	731.256	-.01	4.69	.4511	.0556	-.0133	-.0008	.0006	8.118	.0061	27.0861
114	.935	732.665	-.01	5.93	.5596	.0784	-.0202	.0006	.0004	7.138	.0060	41.7078
115	.935	732.667	-.01	7.15	.6554	.1041	-.0201	.0001	.0005	6.298	.0059	77.0635
116	.935	732.727	-.00	8.31	.7156	.1272	.0021	-.0002	.0003	5.624	.0059	109.8431
117	.936	733.592	-.00	9.54	.7907	.1563	.0087	.0003	.0001	5.058	.0057	131.3556
118	.937	734.470	-.00	10.69	.8557	.1853	.0136	.0007	.0001	4.617	.0056	154.3668
119	.931	729.741	-.01	-.05	-.0024	.0200	.0071	-.0002	.0005	-.120	.0061	10.8584

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 30

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
132	.863	675.122	-.00	-.16	-.0126	.0170	-.0000	.0006	-.0001	.0053	0.0000
133	.863	674.639	-.00	.99	.1006	.0159	.0004	.0006	-.0010	.0053	0.0000
134	.861	673.513	-.01	2.19	.2171	.0120	.0004	.0006	.0004	.0053	0.0000
135	.866	676.976	-.00	3.43	.3458	.0061	.0006	.0006	-.0003	.0053	0.0000
136	.862	673.896	-.00	4.65	.4707	.0011	.0001	.0005	-.0003	.0053	0.0000
137	.862	674.322	-.00	5.86	.5855	.0004	.0005	.0005	-.0001	.0052	0.0000
138	.865	676.724	-.00	7.07	.6594	.0054	-.0015	.0001	.0008	.0052	0.0000
139	.865	676.543	-.00	8.19	.7035	.0087	.0031	.0007	-.0003	.0051	0.0000
140	.865	677.057	-.01	9.31	.7329	.0132	.0095	.0012	-.0001	.0051	0.0000
141	.869	679.740	-.00	10.36	.7202	.0193	.0017	.0003	.0004	.0051	0.0000
142	.865	676.334	-.00	11.45	.7439	.0223	-.0001	.0001	.0006	.0051	0.0000
143	.866	677.540	-.00	12.56	.7846	.0243	.0001	.0000	.0011	.0050	0.0000
144	.869	679.947	-.00	13.76	.8201	.0255	.0001	.0001	.0007	.0050	0.0000
145	.868	679.205	-.00	16.04	.8994	.0263	-.0003	.0002	-.0008	.0047	0.0000
146	.863	674.378	-.01	-.13	-.0144	.0166	.0000	.0006	.0009	.0053	0.0000
147	.863	674.383	-.00	-.13	-.0126	.0169	.0000	.0006	-.0002	.0054	0.0000

TEST= 778 RUN= 30

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
132	.863	675.122	-.00	-.16	-.0126	.0170	.0071	-.0000	.0006	-.736	.0053	9.7798
133	.863	674.639	-.00	.99	.1003	.0177	.0080	.0004	.0006	5.670	.0053	10.2831
134	.861	673.513	-.01	2.19	.2165	.0203	.0110	.0004	.0006	10.672	.0053	12.8000
135	.866	676.976	-.00	3.43	.3449	.0268	.0088	.0006	.0006	12.881	.0053	16.9708
136	.862	673.896	-.00	4.65	.4691	.0393	.0046	.0002	.0005	11.934	.0052	19.6674
137	.862	674.322	-.00	5.86	.5824	.0602	.0007	.0006	.0004	9.669	.0052	38.4718
138	.865	676.724	-.00	7.07	.6537	.0865	.0256	-.0014	.0002	7.558	.0051	75.3856
139	.865	676.543	-.00	8.19	.6951	.1088	.0560	.0031	.0002	6.386	.0051	89.1689
140	.865	677.057	-.01	9.31	.7211	.1316	.0848	.0096	-.0004	5.478	.0050	119.9106
141	.869	679.740	-.00	10.36	.7050	.1485	.1119	.0017	-.0000	4.748	.0051	134.2320
142	.865	676.334	-.00	11.45	.7247	.1695	.1275	-.0000	.0001	4.276	.0050	99.4755
143	.866	677.540	-.00	12.56	.7605	.1944	.1355	.0001	.0000	3.913	.0049	96.3594
144	.869	679.947	-.00	13.76	.7905	.2198	.1423	.0001	.0001	3.596	.0048	108.1047
145	.868	679.205	-.00	16.04	.8572	.2737	.1573	-.0002	.0002	3.132	.0045	104.9886
146	.863	674.378	-.01	-.13	-.0144	.0167	.0076	.0000	.0006	-.863	.0053	10.1633
147	.863	674.383	-.00	-.13	-.0125	.0169	.0065	.0000	.0006	-.741	.0054	10.1633

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 31

					BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
161	.806	624.062	-.01	-.13	-.0138	.0167	-.0002	.0007	.0000	.0049	0.0000
162	.806	623.642	-.01	.98	.0859	.0159	-.0002	.0007	.0018	.0049	0.0000
163	.807	625.147	-.01	2.15	.1931	.0123	.0005	.0008	.0006	.0050	0.0000
164	.807	625.170	-.01	3.34	.2979	.0064	.0001	.0007	.0003	.0049	0.0000
165	.807	625.217	-.01	4.49	.4125	-.0011	.0005	.0007	.0004	.0050	0.0000
166	.807	624.588	-.01	5.73	.5293	-.0075	.0005	.0008	.0009	.0048	0.0000
167	.805	622.862	-.00	6.88	.6035	-.0065	-.0005	.0003	.0011	.0047	0.0000
168	.808	626.038	-.01	8.02	.6591	.0000	.0019	.0007	.0010	.0048	0.0000
169	.807	624.497	-.00	9.12	.6770	.0079	.0009	.0004	.0011	.0048	0.0000
170	.808	625.446	-.00	10.18	.7019	.0135	-.0001	.0002	.0015	.0048	0.0000
171	.808	625.670	-.00	11.31	.7338	.0172	-.0003	.0002	.0011	.0047	0.0000
172	.810	627.917	-.00	12.42	.7652	.0201	-.0002	.0001	.0016	.0047	0.0000
173	.809	626.380	-.00	13.58	.8014	.0224	-.0000	.0002	.0010	.0046	0.0000
174	.811	628.180	-.00	15.81	.8586	.0246	.0001	.0003	.0002	.0044	0.0000
175	.811	628.316	-.00	17.97	.9345	.0254	.0000	.0005	-.0002	.0042	0.0000
176	.812	629.122	.00	20.10	1.0117	.0255	.0000	-.0001	-.0006	.0035	0.0000
177	.806	623.794	-.01	-.17	-.0158	.0167	-.0003	.0007	.0013	.0049	0.0000

TEST= 778 RUN= 31

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
161	.806	624.062	-.01	-.13	-.0137	.0167	.0064	-.0002	.0007	-.820	.0049	8.9887
162	.806	623.642	-.01	.98	.0856	.0173	.0087	-.0002	.0007	4.937	.0049	8.8449
163	.807	625.147	-.01	2.15	.1926	.0195	.0133	.0005	.0008	9.869	.0050	8.5573
164	.807	625.170	-.01	3.34	.2971	.0237	.0195	.0001	.0007	12.520	.0049	10.9303
165	.807	625.217	-.01	4.49	.4113	.0312	.0242	.0005	.0007	13.188	.0050	12.4165
166	.807	624.588	-.01	5.73	.5274	.0453	.0258	.0006	.0007	11.636	.0048	17.3063
167	.805	622.862	-.00	6.88	.5999	.0658	.0431	-.0004	.0003	9.123	.0047	38.1722
168	.808	626.038	-.01	8.02	.6526	.0920	.0771	.0020	.0004	7.094	.0048	61.0036
169	.807	624.497	-.00	9.12	.6672	.1150	.1081	.0009	.0002	5.799	.0047	106.2476
170	.808	625.446	-.00	10.18	.6884	.1374	.1259	-.0001	.0003	5.010	.0047	92.9442
171	.808	625.670	-.00	11.31	.7162	.1609	.1412	-.0002	.0002	4.452	.0046	96.5397
172	.810	627.917	-.00	12.42	.7430	.1842	.1543	-.0002	.0002	4.033	.0046	106.6072
173	.809	626.380	-.00	13.58	.7737	.2099	.1658	.0000	.0002	3.686	.0045	90.2476
174	.811	628.180	-.00	15.81	.8194	.2576	.1778	.0002	.0002	3.181	.0043	85.3936
175	.811	628.316	-.00	17.97	.8810	.3124	.1830	.0002	.0005	2.820	.0040	88.2700
176	.812	629.122	.00	20.10	.9413	.3716	.1847	-.0000	-.0001	2.533	.0033	105.1690
177	.806	623.794	-.01	-.17	-.0158	.0167	.0062	-.0003	.0007	-.944	.0049	9.1805

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 32

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
191	.753	573.003	-.01	-.10	-.0130	.0165	-.0001	.0008	.0012	.0047	0.0000
192	.752	572.404	-.01	.97	.0817	.0158	-.0000	.0007	.0018	.0047	0.0000
193	.755	575.026	-.01	2.11	.1816	.0125	.0002	.0008	.0011	.0048	0.0000
194	.754	573.551	-.01	3.24	.2786	.0070	.0002	.0007	.0004	.0047	0.0000
195	.754	574.187	-.01	4.38	.3724	.0004	.0001	.0007	.0012	.0047	0.0000
196	.754	574.389	-.01	5.57	.4708	-.0032	.0005	.0007	.0014	.0047	0.0000
197	.755	575.239	-.01	6.72	.5439	.0010	.0018	.0010	.0004	.0047	0.0000
198	.756	576.110	-.00	7.84	.5951	.0060	.0006	.0005	.0005	.0047	0.0000
199	.754	574.272	-.01	8.96	.6356	.0097	.0005	.0005	.0023	.0046	0.0000
200	.756	575.603	-.00	10.09	.6831	.0126	.0004	.0004	.0017	.0046	0.0000
201	.755	575.139	-.00	11.21	.7320	.0153	-.0009	.0003	.0011	.0045	0.0000
202	.755	574.965	-.00	12.30	.7612	.0175	-.0011	.0003	.0021	.0044	0.0000
203	.756	575.774	-.00	13.44	.7837	.0194	-.0006	.0003	.0010	.0044	0.0000
204	.757	576.908	-.00	15.67	.8412	.0224	.0002	.0004	.0008	.0043	0.0000
205	.758	577.467	-.00	17.80	.9052	.0236	.0002	.0005	.0003	.0041	0.0000
206	.758	577.881	-.00	19.92	.9799	.0249	.0002	.0000	.0025	.0034	0.0000
207	.758	577.645	-.00	22.02	1.0588	.0238	-.0003	.0004	-.0013	.0027	0.0000
210	.754	573.991	-.01	-.14	-.0144	.0165	.0007	.0008	.0017	.0047	0.0000
211	.754	573.836	-.01	.97	.0814	.0159	-.0002	.0008	.0020	.0048	0.0000
212	.754	573.397	-.01	2.10	.1761	.0125	.0003	.0008	.0023	.0047	0.0000
213	.754	573.413	-.01	3.24	.2784	.0071	.0001	.0007	.0017	.0047	0.0000
214	.753	573.004	-.01	4.40	.3731	.0003	.0000	.0007	.0012	.0047	0.0000
215	.753	572.362	-.01	5.57	.4703	-.0032	.0005	.0007	.0027	.0047	0.0000

TEST= 778 RUN= 32

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
191	.753	573.003	-.01	-.10	-.0129	.0165	.0057	-.0001	.0008	-.783	.0047	8.3416
192	.752	572.404	-.01	.97	.0814	.0172	.0096	-.0000	.0007	4.726	.0047	8.6652
193	.755	575.026	-.01	2.11	.1810	.0192	.0149	.0002	.0008	9.426	.0048	8.3296
194	.754	573.551	-.01	3.24	.2777	.0228	.0219	.0003	.0007	12.181	.0047	8.7011
195	.754	574.187	-.01	4.38	.3713	.0288	.0279	.0001	.0006	12.884	.0047	9.6120
196	.754	574.389	-.01	5.57	.4688	.0425	.0337	.0005	.0006	11.025	.0047	15.6045
197	.755	575.239	-.01	6.72	.5401	.0646	.0575	.0019	.0008	8.356	.0047	45.1834
198	.756	576.110	-.00	7.84	.5888	.0871	.0854	.0007	.0004	6.760	.0046	66.3969
199	.754	574.272	-.01	8.96	.6264	.1086	.1100	.0006	.0004	5.768	.0045	81.2588
200	.756	575.603	-.00	10.09	.6704	.1321	.1287	.0004	.0003	5.075	.0045	106.4274
201	.755	575.139	-.00	11.21	.7151	.1574	.1441	-.0009	.0005	4.544	.0044	209.9772
202	.755	574.965	-.00	12.30	.7400	.1792	.1566	-.0010	.0005	4.129	.0043	257.0790
203	.756	575.774	-.00	13.44	.7577	.2010	.1713	-.0005	.0004	3.769	.0043	139.8653
204	.757	576.908	-.00	15.67	.8039	.2487	.1868	.0003	.0003	3.232	.0042	77.3037
205	.758	577.467	-.00	17.80	.8547	.2992	.1946	.0003	.0004	2.856	.0039	81.7981
206	.758	577.881	-.00	19.92	.9128	.3572	.1926	.0002	-.0001	2.555	.0032	85.7532
207	.758	577.645	-.00	22.02	.9727	.4190	.2004	-.0001	.0005	2.322	.0025	98.3375
210	.754	573.991	-.01	-.14	-.0144	.0166	.0055	.0007	.0008	-.867	.0047	8.4374
211	.754	573.836	-.01	.97	.0811	.0172	.0096	-.0001	.0008	4.703	.0048	8.3895
212	.754	573.397	-.01	2.10	.1756	.0189	.0135	.0003	.0008	9.286	.0047	8.1498
213	.754	573.413	-.01	3.24	.2776	.0228	.0219	.0002	.0007	12.165	.0047	8.7970
214	.753	573.004	-.01	4.40	.3720	.0290	.0272	.0001	.0007	12.849	.0047	8.9168
215	.753	572.362	-.01	5.57	.4684	.0424	.0343	.0005	.0007	11.037	.0047	15.0771

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 33

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
229	.700	519.721	-.01	-.08	-.0183	.0165	.0004	.0008	.0009	.0045	0.0000
230	.701	520.739	-.01	.97	.0721	.0160	-.0002	.0007	.0010	.0046	0.0000
231	.701	520.357	-.00	2.09	.1652	.0127	-.0002	.0007	.0003	.0046	0.0000
232	.702	520.835	-.00	3.23	.2538	.0076	.0002	.0006	-.0005	.0046	0.0000
233	.702	521.323	-.00	4.32	.3459	.0011	-.0003	.0005	.0007	.0045	0.0000
234	.702	521.404	-.00	5.50	.4411	-.0034	.0002	.0005	.0008	.0045	0.0000
235	.701	520.658	-.00	6.63	.5189	.0002	.0010	.0007	-.0008	.0045	0.0000
236	.703	521.963	-.00	7.74	.5704	.0053	-.0004	.0003	.0010	.0044	0.0000
237	.703	522.247	-.00	8.87	.6198	.0082	.0002	.0004	.0008	.0044	0.0000
238	.702	521.897	-.00	9.96	.6633	.0102	.0004	.0004	.0013	.0043	0.0000
239	.702	521.566	-.00	11.09	.7001	.0127	.0001	.0003	.0009	.0043	0.0000
240	.703	522.641	-.00	12.18	.7351	.0149	-.0001	.0003	.0006	.0043	0.0000
241	.703	522.496	-.00	13.33	.7697	.0169	-.0001	.0002	.0014	.0042	0.0000
242	.704	523.378	-.00	15.56	.8330	.0206	.0000	.0004	.0005	.0041	0.0000
243	.705	524.332	-.00	17.68	.8896	.0219	-.0001	.0004	.0001	.0040	0.0000
244	.705	524.174	-.00	19.77	.9555	.0243	.0001	.0003	-.0003	.0034	0.0000
245	.706	525.533	.00	21.87	1.0225	.0235	.0002	.0002	-.0009	.0027	0.0000
246	.699	518.136	-.01	-.11	-.0184	.0164	.0002	.0008	.0011	.0044	0.0000

TEST= 778 RUN= 33

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
229	.700	519.721	-.01	-.08	-.0183	.0165	.0051	.0004	.0009	-1.106	.0045	8.7730
230	.701	520.739	-.01	.97	.0719	.0172	.0094	-.0002	.0007	4.185	.0046	7.5505
231	.701	520.357	-.00	2.09	.1646	.0187	.0149	-.0001	.0007	8.792	.0046	7.7663
232	.702	520.835	-.00	3.23	.2530	.0219	.0214	.0002	.0006	11.546	.0046	7.7663
233	.702	521.323	-.00	4.32	.3448	.0271	.0284	-.0003	.0005	12.722	.0045	8.0539
234	.702	521.404	-.00	5.50	.4394	.0388	.0372	.0003	.0005	11.314	.0045	10.1633
235	.701	520.658	-.00	6.63	.5154	.0601	.0545	.0011	.0006	8.574	.0045	42.0074
236	.703	521.963	-.00	7.74	.5644	.0821	.0846	-.0004	.0003	6.875	.0044	57.6478
237	.703	522.247	-.00	8.87	.6111	.1037	.1106	.0003	.0004	5.895	.0043	67.1160
238	.702	521.897	-.00	9.96	.6516	.1248	.1303	.0004	.0003	5.222	.0043	69.7527
239	.702	521.566	-.00	11.09	.6846	.1471	.1478	.0002	.0003	4.653	.0042	75.8650
240	.703	522.641	-.00	12.18	.7154	.1697	.1617	-.0001	.0003	4.216	.0042	74.7864
241	.703	522.496	-.00	13.33	.7450	.1940	.1743	-.0000	.0002	3.840	.0041	74.4268
242	.704	523.378	-.00	15.56	.7969	.2433	.1931	.0001	.0004	3.275	.0039	75.2658
243	.705	524.332	-.00	17.68	.8409	.2911	.2015	-.0000	.0004	2.889	.0038	73.8276
244	.705	524.174	-.00	19.77	.8910	.3460	.1977	.0001	.0002	2.575	.0032	84.0148
245	.706	525.533	.00	21.87	.9401	.4026	.2022	.0003	.0001	2.335	.0025	92.4043
246	.699	518.136	-.01	-.11	-.0184	.0165	.0043	.0002	.0008	-1.116	.0044	7.7903

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 34

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
260	.496	305.654	-.00	-.01	-.0118	.0171	.0004	.0007	.0019	.0041	0.0000
261	.497	306.717	-.00	1.01	.0667	.0166	.0000	.0008	.0017	.0041	0.0000
262	.496	306.097	-.00	2.04	.1454	.0135	-.0004	.0008	.0016	.0041	0.0000
263	.498	307.253	-.00	3.09	.2280	.0095	.0004	.0007	.0003	.0041	0.0000
264	.498	307.786	-.00	4.11	.3022	.0029	-.0001	.0006	.0002	.0040	0.0000
265	.499	308.416	-.00	5.20	.3800	-.0045	-.0001	.0002	.0020	.0041	0.0000
266	.498	307.846	-.00	6.32	.4704	-.0067	.0001	.0002	.0012	.0040	0.0000
267	.498	307.310	-.00	7.40	.5370	-.0024	-.0003	.0004	.0008	.0040	0.0000
268	.497	306.775	-.00	8.50	.5933	.0008	.0005	.0005	-.0010	.0040	0.0000
269	.497	306.855	.00	9.56	.6404	.0022	.0004	.0003	-.0018	.0039	0.0000
270	.499	308.538	-.00	10.68	.6872	.0037	.0000	.0004	-.0000	.0038	0.0000
271	.497	306.673	-.00	11.72	.7200	.0059	.0001	.0005	.0017	.0037	0.0000
272	.499	308.552	-.00	12.86	.7631	.0081	.0002	.0002	.0009	.0037	0.0000
273	.498	307.425	-.00	15.05	.8272	.0119	-.0000	.0002	.0002	.0035	0.0000
274	.498	307.463	-.00	17.15	.8923	.0151	-.0003	.0003	-.0004	.0035	0.0000
275	.497	306.976	-.00	19.24	.9524	.0175	-.0001	.0003	-.0012	.0032	0.0000
276	.498	307.828	-.00	21.24	1.0004	.0206	-.0009	.0006	-.0004	.0025	0.0000
277	.496	305.290	-.00	-.07	-.0160	.0173	-.0001	.0007	.0024	.0040	0.0000

TEST= 778 RUN= 34

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
260	.496	305.654	-.00	-.01	-.0118	.0171	.0053	.0004	.0007	-.692	.0041	4.5663
261	.497	306.717	-.00	1.01	.0664	.0177	.0111	.0000	.0008	3.745	.0041	4.7101
262	.496	306.097	-.00	2.04	.1449	.0187	.0170	-.0004	.0009	7.761	.0041	4.6622
263	.498	307.253	-.00	3.09	.2272	.0218	.0233	.0004	.0007	10.433	.0041	4.7700
264	.498	307.786	-.00	4.11	.3012	.0246	.0286	-.0000	.0006	12.246	.0040	4.3985
265	.499	308.416	-.00	5.20	.3788	.0300	.0357	-.0001	.0002	12.646	.0041	4.4344
266	.498	307.846	-.00	6.32	.4683	.0451	.0442	.0001	.0002	10.389	.0040	9.4202
267	.498	307.310	-.00	7.40	.5328	.0667	.0621	-.0003	.0004	7.987	.0040	32.0599
268	.497	306.775	-.00	8.50	.5866	.0885	.0884	.0006	.0005	6.630	.0039	51.8950
269	.497	306.855	.00	9.56	.6312	.1086	.1136	.0005	.0003	5.814	.0038	55.4905
270	.499	308.538	-.00	10.68	.6746	.1310	.1384	.0001	.0004	5.149	.0037	56.4493
271	.497	306.673	-.00	11.72	.7038	.1520	.1557	.0002	.0005	4.629	.0036	60.4044
272	.499	308.552	-.00	12.86	.7422	.1777	.1720	.0002	.0001	4.176	.0036	61.2433
273	.498	307.425	-.00	15.05	.7958	.2262	.1983	.0000	.0002	3.517	.0034	67.3557
274	.498	307.463	-.00	17.15	.8482	.2776	.2190	-.0002	.0003	3.056	.0033	62.4418
275	.497	306.976	-.00	19.24	.8935	.3303	.2284	-.0000	.0003	2.705	.0030	62.3220
276	.498	307.828	-.00	21.24	.9250	.3817	.2195	-.0006	.0009	2.423	.0024	71.5504
277	.496	305.290	-.00	-.07	-.0160	.0174	.0049	-.0001	.0007	-.921	.0040	4.3386

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 35

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
278	.290	116.858	-.00	-.07	-.0209	.0178	-.0006	.0008	.0005	.0038	0.0000
279	.291	117.535	-.00	.94	.0580	.0176	.0001	.0011	-.0006	.0039	0.0000
280	.292	118.114	-.00	1.93	.1358	.0148	.0001	.0008	-.0012	.0039	0.0000
281	.293	118.696	-.00	2.97	.2028	.0120	-.0003	.0006	-.0013	.0039	0.0000
282	.294	119.373	-.00	3.95	.2684	.0066	-.0003	.0009	.0041	.0038	0.0000
283	.292	118.024	-.00	4.99	.3500	-.0006	.0002	.0007	.0033	.0039	0.0000
284	.291	117.355	-.00	6.06	.4206	-.0070	.0002	.0005	.0028	.0039	0.0000
285	.291	117.377	-.00	7.08	.4971	-.0054	-.0003	.0003	.0027	.0038	0.0000
286	.292	117.981	-.00	8.14	.5607	-.0032	.0001	.0005	.0014	.0038	0.0000
287	.292	117.909	-.00	9.18	.6179	-.0027	.0007	.0007	.0000	.0036	0.0000
288	.291	117.258	-.00	10.26	.6579	-.0020	.0007	.0008	-.0008	.0037	0.0000
289	.292	117.963	-.00	11.32	.7009	-.0006	.0012	.0005	-.0020	.0035	0.0000
290	.292	117.994	-.00	12.41	.7462	.0008	.0000	.0007	-.0017	.0035	0.0000
291	.291	117.385	.00	14.57	.8238	.0048	-.0005	.0005	-.0024	.0033	0.0000
292	.292	118.034	.00	16.63	.8835	.0086	-.0005	.0002	-.0035	.0031	0.0000
293	.289	116.060	.00	18.67	.9145	.0111	-.0007	.0003	-.0036	.0028	0.0000
294	.292	118.086	.00	20.69	1.0026	.0145	-.0001	.0006	-.0051	.0025	0.0000
295	.291	117.064	-.00	10.26	.6591	-.0021	.0007	.0008	-.0007	.0035	0.0000

TEST= 778 RUN= 35

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
278	.290	116.858	-.00	-.07	-.0209	.0178	.0044	-.0006	.0008	-1.172	.0038	1.5940
279	.291	117.535	-.00	.94	.0577	.0186	.0091	.0001	.0010	3.107	.0039	1.7738
280	.292	118.114	-.00	1.93	.1352	.0194	.0172	.0001	.0008	6.980	.0039	1.8337
281	.293	118.696	-.00	2.97	.2019	.0225	.0241	-.0003	.0006	8.985	.0039	1.9296
282	.294	119.373	-.00	3.95	.2673	.0251	.0309	-.0002	.0010	10.659	.0038	2.1933
283	.292	118.024	-.00	4.99	.3488	.0299	.0359	.0002	.0007	11.680	.0039	1.6180
284	.291	117.355	-.00	6.06	.4190	.0374	.0432	.0002	.0005	11.196	.0039	2.6727
285	.291	117.377	-.00	7.08	.4939	.0559	.0511	-.0003	.0004	8.843	.0037	9.9955
286	.292	117.981	-.00	8.14	.5555	.0763	.0717	.0002	.0005	7.283	.0037	20.6142
287	.292	117.909	-.00	9.18	.6104	.0959	.0915	.0008	.0005	6.367	.0036	28.7640
288	.291	117.258	-.00	10.26	.6478	.1152	.1166	.0009	.0006	5.621	.0036	32.7790
289	.292	117.963	-.00	11.32	.6874	.1370	.1381	.0012	.0002	5.019	.0034	30.2022
290	.292	117.994	-.00	12.41	.7286	.1611	.1567	.0002	.0006	4.522	.0034	33.9775
291	.291	117.385	.00	14.57	.7961	.2118	.1903	-.0003	.0006	3.758	.0032	38.2321
292	.292	118.034	.00	16.63	.8440	.2611	.2175	-.0004	.0004	3.232	.0030	34.8764
293	.289	116.060	.00	18.67	.8628	.3032	.2299	-.0006	.0005	2.846	.0026	30.7415
294	.292	118.086	.00	20.69	.9329	.3677	.2433	.0001	.0006	2.537	.0023	31.5805
295	.291	117.064	-.00	10.26	.6490	.1154	.1133	.0009	.0006	5.623	.0035	33.3183

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 36

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
357	.932	736.432	-.01	-.13	-.0011	.0264	.0017	.0004	.0012	.0061	0.0000
358	.931	735.885	-.01	1.04	.1043	.0266	.0020	.0004	.0015	.0061	0.0000
359	.933	737.326	-.00	2.25	.1987	.0294	.0032	.0003	.0008	.0061	0.0000
360	.933	737.090	-.01	3.43	.3045	.0289	.0024	.0004	.0020	.0061	0.0000
361	.934	738.256	-.01	4.61	.4070	.0294	.0022	.0004	.0016	.0061	0.0000
362	.934	737.675	-.00	5.83	.5007	.0294	.0016	.0004	.0007	.0059	0.0000
363	.936	739.255	-.00	7.08	.6060	.0306	.0024	.0002	.0014	.0060	0.0000
364	.936	739.544	-.01	8.26	.6861	.0317	.0018	.0006	.0006	.0058	0.0000
365	.938	740.830	-.01	9.46	.7613	.0324	.0029	.0006	-.0000	.0057	0.0000
366	.940	742.423	-.01	10.64	.8299	.0321	.0018	.0005	.0006	.0057	0.0000
367	.942	744.413	-.01	11.92	.9452	.0334	.0048	.0007	-.0000	.0053	0.0000
368	.932	736.560	-.01	-.06	-.0072	.0273	.0019	.0005	.0011	.0061	0.0000

TEST= 778 RUN= 36

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
357	.932	736.432	-.01	-.13	-.0010	.0264	.0094	.0017	.0004	-.038	.0061	18.3905
358	.931	735.885	-.01	1.04	.1038	.0285	.0064	.0020	.0004	3.646	.0061	50.7586
359	.933	737.326	-.00	2.25	.1974	.0372	.0134	.0032	.0002	5.312	.0061	49.3793
360	.933	737.090	-.01	3.43	.3022	.0470	.0127	.0024	.0002	6.424	.0060	45.7011
361	.934	738.256	-.01	4.61	.4033	.0620	.0129	.0023	.0002	6.506	.0061	44.2299
362	.934	737.675	-.00	5.83	.4951	.0801	.0164	.0017	.0002	6.181	.0059	60.8735
363	.936	739.255	-.00	7.08	.5976	.1050	.0169	.0024	-.0000	5.693	.0060	125.2413
364	.936	739.544	-.01	8.26	.6744	.1299	.0157	.0019	.0003	5.190	.0058	156.6896
365	.938	740.830	-.01	9.46	.7457	.1571	.0235	.0030	.0001	4.748	.0057	139.5862
366	.940	742.423	-.01	10.64	.8097	.1847	.0284	.0018	.0002	4.384	.0056	143.4482
367	.942	744.413	-.01	11.92	.9179	.2278	.0006	.0048	-.0003	4.029	.0052	156.6896
368	.932	736.560	-.01	-.06	-.0072	.0273	.0092	.0019	.0005	-.263	.0061	16.8089

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 77d RUN= 37

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
418	.863	684.124	-.01	.04	-.0198	.0166	.0019	.0006	.0013	.0054	0.0000
419	.862	682.995	-.01	1.15	.1030	.0154	.0014	.0006	.0022	.0054	0.0000
420	.866	686.530	-.01	2.40	.2354	.0121	.0013	.0007	.0017	.0054	0.0000
421	.862	682.905	-.01	3.63	.3605	.0075	.0015	.0005	.0022	.0053	0.0000
422	.864	684.882	-.01	4.72	.4730	.0076	.0023	.0004	.0021	.0053	0.0000
423	.864	685.422	-.01	5.88	.5583	.0092	.0057	.0012	.0003	.0052	0.0000
424	.864	685.225	-.01	7.06	.6370	.0117	.0014	.0006	.0011	.0052	0.0000
425	.866	686.793	-.01	8.17	.6789	.0132	.0081	.0016	-.0005	.0052	0.0000
426	.864	684.830	-.01	9.30	.7242	.0129	.0066	.0017	.0001	.0051	0.0000
427	.865	685.870	-.01	10.35	.7336	.0172	.0090	.0018	-.0008	.0051	0.0000
428	.862	683.664	-.01	11.42	.7316	.0215	.0091	.0019	-.0012	.0051	0.0000
429	.862	683.316	-.01	-.21	-.0237	.0163	.0015	.0007	.0027	.0053	0.0000

TEST= 77d RUN= 37

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
418	.863	684.124	-.01	.04	-.0198	.0166	.0104	.0019	.0006	-1.196	.0054	12.9837
419	.862	682.995	-.01	1.15	.1027	.0175	.0119	.0014	.0006	5.878	.0054	12.5791
420	.866	686.530	-.01	2.40	.2347	.0219	.0078	.0013	.0006	10.713	.0054	17.3239
421	.862	682.905	-.01	3.63	.3593	.0303	.0021	.0016	.0004	11.863	.0053	19.4939
422	.864	684.882	-.01	4.72	.4708	.0465	.0014	.0023	.0002	10.125	.0053	28.6896
423	.864	685.422	-.01	5.88	.5544	.0663	.0162	.0058	.0006	8.356	.0052	105.9316
424	.864	685.225	-.01	7.06	.6307	.0898	.0387	.0014	.0004	7.020	.0051	65.4716
425	.866	686.793	-.01	8.17	.6702	.1095	.0578	.0082	.0005	6.121	.0051	91.9545
426	.864	684.830	-.01	9.30	.7126	.1298	.0766	.0068	.0006	5.492	.0050	106.4833
427	.865	685.870	-.01	10.35	.7186	.1487	.0986	.0092	.0002	4.833	.0050	124.3225
428	.862	683.664	-.01	11.42	.7129	.1659	.1141	.0093	.0001	4.297	.0050	120.0000
429	.862	683.316	-.01	-.21	-.0236	.0164	.0100	.0015	.0007	-1.439	.0053	13.6458

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 38

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
444	.806	632.344	-.01	-.18	-.0255	.0165	.0012	.0006	.0011	.0049	0.0000
445	.806	632.418	-.01	.99	.0820	.0154	.0013	.0006	.0015	.0049	0.0000
446	.806	632.571	-.01	2.13	.1951	.0114	.0013	.0006	.0009	.0049	0.0000
447	.806	632.732	-.00	3.37	.3100	.0042	.0013	.0005	.0004	.0049	0.0000
448	.804	631.353	-.01	4.54	.4239	-.0037	.0012	.0005	.0010	.0049	0.0000
449	.806	633.118	-.00	5.76	.5405	-.0075	.0004	.0004	.0011	.0049	0.0000
450	.806	633.351	-.00	6.93	.6051	-.0050	.0012	.0005	.0005	.0048	0.0000
451	.806	633.021	-.01	8.06	.6584	-.0024	.0067	.0018	-.0008	.0048	0.0000
452	.806	633.329	-.01	9.17	.6920	.0033	.0075	.0018	-.0013	.0047	0.0000
453	.808	634.548	-.00	10.24	.7374	.0063	-.0019	.0005	.0004	.0047	0.0000
454	.808	634.378	-.01	11.36	.7419	.0142	.0078	.0018	-.0027	.0046	0.0000
455	.808	634.984	-.00	12.43	.7447	.0206	.0015	.0007	-.0002	.0046	0.0000
456	.808	635.018	-.00	13.57	.7672	.0239	.0005	.0005	-.0007	.0046	0.0000
457	.810	636.461	-.00	15.82	.8334	.0266	.0002	.0008	-.0012	.0044	0.0000
458	.810	636.770	-.01	18.01	.9107	.0276	.0006	.0014	-.0032	.0041	0.0000
459	.812	638.686	-.00	20.11	.9863	.0289	.0001	.0014	-.0064	.0035	0.0000
460	.815	640.771	.00	22.32	1.0944	.0273	-.0006	.0016	-.0099	.0026	0.0000
461	.807	633.846	-.00	5.83	.5490	-.0067	.0004	.0003	.0010	.0049	0.0000

TEST= 778 RUN= 38

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
444	.806	632.344	-.01	-.18	-.0255	.0166	.0089	.0012	.0006	-1.534	.0049	12.0642
445	.806	632.418	-.01	.99	.0817	.0168	.0129	.0013	.0006	4.856	.0049	12.4320
446	.806	632.571	-.01	2.13	.1945	.0186	.0155	.0014	.0006	10.443	.0049	12.5055
447	.806	632.732	-.00	3.37	.3092	.0224	.0164	.0013	.0004	13.786	.0049	12.8733
448	.804	631.353	-.01	4.54	.4228	.0298	.0191	.0012	.0004	14.171	.0049	16.5147
449	.806	633.118	-.00	5.76	.5385	.0467	.0242	.0004	.0004	11.526	.0048	30.7860
450	.806	633.351	-.00	6.93	.6013	.0681	.0502	.0012	.0004	8.826	.0048	56.3678
451	.806	633.021	-.01	8.06	.6522	.0899	.0717	.0069	.0009	7.257	.0047	78.3452
452	.806	633.329	-.01	9.17	.6826	.1136	.0949	.0077	.0006	6.009	.0047	102.9890
453	.808	634.548	-.00	10.24	.7245	.1373	.0888	-.0018	.0009	5.277	.0046	124.1386
454	.808	634.378	-.01	11.36	.7246	.1600	.1250	.0080	.0002	4.530	.0046	187.5841
455	.808	634.984	-.00	12.43	.7228	.1805	.1590	.0016	.0003	4.005	.0045	160.3660
456	.808	635.018	-.00	13.57	.7402	.2032	.1686	.0006	.0004	3.642	.0045	105.5620
457	.810	636.461	-.00	15.82	.7945	.2527	.1840	.0005	.0007	3.144	.0042	101.1499
458	.810	636.770	-.01	18.01	.8575	.3079	.1878	.0010	.0011	2.785	.0039	104.0925
459	.812	638.686	-.00	20.11	.9162	.3663	.1847	.0006	.0013	2.501	.0033	124.3205
460	.815	640.771	.00	22.32	1.0020	.4408	.1950	.0001	.0017	2.273	.0024	157.0557
461	.807	633.846	-.00	5.83	.5469	.0492	.0239	.0004	.0003	11.124	.0048	30.8963

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 39

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
475	.754	582.493	-.01	-.12	-.0239	.0164	.0010	.0006	.0015	.0047	0.0000
476	.754	582.494	-.01	.96	.0732	.0156	.0014	.0006	.0018	.0047	0.0000
477	.753	581.527	-.01	2.13	.1767	.0119	.0015	.0005	.0011	.0047	0.0000
478	.754	582.733	-.00	3.28	.2810	.0052	.0013	.0005	.0007	.0047	0.0000
479	.753	581.705	-.00	4.43	.3820	-.0029	.0005	.0003	.0008	.0047	0.0000
480	.756	584.456	-.00	5.63	.4809	-.0116	.0006	.0003	.0013	.0047	0.0000
481	.754	582.645	-.01	6.80	.5717	-.0122	.0028	.0013	-.0002	.0047	0.0000
482	.755	583.947	-.01	7.96	.6349	-.0101	.0069	.0020	-.0011	.0046	0.0000
483	.753	581.611	-.01	9.07	.6744	-.0052	.0079	.0021	-.0017	.0045	0.0000
484	.754	582.969	-.01	10.12	.7057	.0013	.0086	.0022	-.0019	.0045	0.0000
485	.756	584.816	-.01	11.27	.7171	.0105	.0017	.0007	.0009	.0045	0.0000
486	.756	585.279	-.00	12.34	.7444	.0152	.0004	.0005	.0002	.0044	0.0000
487	.755	584.149	-.00	13.46	.7630	.0188	.0003	.0004	.0012	.0044	0.0000
488	.755	584.418	-.00	15.70	.8249	.0234	.0000	.0006	-.0006	.0043	0.0000
489	.757	585.719	-.01	17.85	.8835	.0255	.0000	.0014	-.0035	.0041	0.0000
490	.757	585.931	-.00	19.97	.9590	.0273	.0002	.0015	-.0039	.0034	0.0000
491	.761	589.283	.00	22.10	1.0496	.0265	-.0005	.0015	-.0088	.0027	0.0000
492	.755	583.410	-.01	-.12	-.0255	.0164	.0017	.0006	.0021	.0047	0.0000

TEST= 778 RUN= 39

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
475	.754	582.493	-.01	-.12	-.0239	.0165	.0079	.0010	.0006	-1.449	.0047	10.6288
476	.754	582.494	-.01	.96	.0730	.0168	.0129	.0014	.0006	4.336	.0047	13.2779
477	.753	581.527	-.01	2.13	.1761	.0184	.0158	.0015	.0005	9.547	.0047	10.4826
478	.754	582.733	-.00	3.28	.2802	.0213	.0209	.0013	.0004	13.142	.0047	10.8136
479	.753	581.705	-.00	4.43	.3811	.0266	.0270	.0005	.0003	14.336	.0047	11.3653
480	.756	584.456	-.00	5.63	.4797	.0356	.0309	.0006	.0002	13.464	.0047	14.7860
481	.754	582.645	-.01	6.80	.5691	.0555	.0457	.0029	.0010	10.254	.0046	37.3333
482	.755	583.947	-.01	7.96	.6302	.0779	.0668	.0071	.0010	8.094	.0046	60.5977
483	.753	581.611	-.01	9.07	.6668	.1012	.0924	.0081	.0008	6.586	.0045	77.6096
484	.754	582.969	-.01	10.12	.6945	.1253	.1128	.0089	.0007	5.543	.0045	100.2304
485	.756	584.816	-.01	11.27	.7012	.1505	.1499	.0018	.0004	4.659	.0044	180.5957
486	.756	585.279	-.00	12.34	.7239	.1739	.1669	.0005	.0004	4.162	.0043	110.8971
487	.755	584.149	-.00	13.46	.7377	.1959	.1782	.0004	.0003	3.765	.0043	97.6557
488	.755	584.418	-.00	15.70	.7878	.2457	.1961	.0002	.0006	3.206	.0041	90.6671
489	.757	585.719	-.01	17.85	.8331	.2952	.2008	.0004	.0014	2.823	.0039	86.6211
490	.757	585.931	-.00	19.97	.8920	.3532	.1997	.0007	.0013	2.526	.0032	98.5752
491	.761	589.283	.00	22.10	.9625	.4194	.2055	.0001	.0016	2.295	.0025	118.2535
492	.755	583.410	-.01	-.12	-.0254	.0165	.0084	.0017	.0006	-1.543	.0047	10.8127

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 40

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
506	.702	530.006	-.01	-.11	-.0216	.0164	.0015	.0007	.0013	.0045	0.0000
507	.701	529.460	-.01	.98	.0743	.0156	.0018	.0006	.0018	.0045	0.0000
508	.701	529.543	-.01	2.10	.1697	.0122	.0013	.0006	.0017	.0045	0.0000
509	.702	530.722	-.00	3.26	.2692	.0059	.0012	.0006	.0012	.0045	0.0000
510	.702	529.719	-.00	4.34	.3588	-.0018	.0010	.0004	.0008	.0045	0.0000
511	.701	529.360	-.00	5.52	.4522	-.0114	.0006	.0004	.0018	.0045	0.0000
512	.703	531.649	-.00	6.67	.5431	-.0170	.0007	.0006	.0010	.0045	0.0000
513	.701	529.215	-.01	7.82	.6052	-.0122	.0045	.0018	-.0004	.0044	0.0000
514	.705	533.688	-.01	8.92	.6484	-.0051	.0079	.0022	-.0018	.0044	0.0000
515	.701	529.537	-.01	9.99	.6853	.0004	.0087	.0025	-.0019	.0043	0.0000
516	.702	529.880	-.01	11.12	.7158	.0068	.0044	.0012	-.0004	.0043	0.0000
517	.703	531.056	-.00	12.22	.7426	.0109	.0002	.0005	.0013	.0042	0.0000
518	.704	531.934	-.00	13.36	.7688	.0143	-.0001	.0004	.0010	.0042	0.0000
519	.704	532.438	-.00	15.58	.8215	.0202	-.0001	.0007	-.0012	.0041	0.0000
520	.703	531.597	-.00	17.70	.8807	.0230	-.0000	.0013	-.0031	.0039	0.0000
521	.704	532.388	-.01	19.81	.9432	.0260	-.0001	.0016	-.0032	.0034	0.0000
522	.706	533.963	-.01	21.86	.9997	.0257	.0004	.0014	-.0026	.0028	0.0000
523	.702	530.228	-.01	-.11	-.0196	.0164	.0014	.0006	.0027	.0045	0.0000

TEST= 778 RUN= 40

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
506	.702	530.006	-.01	-.11	-.0216	.0165	.0084	.0015	.0007	-1.309	.0045	11.2357
507	.701	529.460	-.01	.98	.0740	.0168	.0135	.0018	.0006	4.397	.0045	11.5667
508	.701	529.543	-.01	2.10	.1691	.0184	.0177	.0013	.0006	9.199	.0045	11.3285
509	.702	530.722	-.00	3.26	.2684	.0212	.0217	.0012	.0005	12.660	.0045	11.2182
510	.702	529.719	-.00	4.34	.3579	.0254	.0284	.0010	.0004	14.112	.0045	11.2550
511	.701	529.360	-.00	5.52	.4512	.0322	.0363	.0006	.0003	14.031	.0045	10.0412
512	.703	531.649	-.00	6.67	.5414	.0462	.0440	.0008	.0005	11.711	.0044	16.2204
513	.701	529.215	-.01	7.82	.6012	.0703	.0647	.0047	.0012	8.555	.0044	50.7586
514	.705	533.688	-.01	8.92	.6413	.0955	.0919	.0081	.0010	6.714	.0043	74.2992
515	.701	529.537	-.01	9.99	.6749	.1192	.1116	.0090	.0010	5.661	.0042	91.5867
516	.702	529.880	-.01	11.12	.7010	.1448	.1414	.0046	.0004	4.842	.0042	222.5263
517	.703	531.056	-.00	12.22	.7234	.1679	.1691	.0003	.0005	4.309	.0041	100.4127
518	.704	531.934	-.00	13.36	.7447	.1916	.1826	.0000	.0004	3.887	.0041	94.5292
519	.704	532.438	-.00	15.58	.7859	.2400	.2021	.0001	.0006	3.275	.0040	93.2419
520	.703	531.597	-.00	17.70	.8320	.2896	.2109	.0004	.0012	2.873	.0037	84.4142
521	.704	532.388	-.01	19.81	.8785	.3441	.2081	.0005	.0015	2.553	.0032	90.8510
522	.706	533.963	-.01	21.86	.9182	.3962	.2076	.0009	.0011	2.318	.0026	109.9776
523	.702	530.228	-.01	-.11	-.0196	.0164	.0086	.0014	.0007	-1.191	.0045	10.6288

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 41

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
537	.497	312.041	-.01	-.05	-.0226	.0169	.0008	.0007	.0038	.0040	0.0000
538	.498	313.282	-.00	.98	.0623	.0163	.0011	.0006	.0029	.0040	0.0000
539	.497	312.222	-.00	2.00	.1399	.0136	.0009	.0006	.0027	.0040	0.0000
540	.498	313.463	-.00	3.09	.2243	.0085	.0015	.0005	.0017	.0040	0.0000
541	.498	313.465	-.00	4.12	.3008	.0018	.0004	.0003	.0021	.0040	0.0000
542	.499	314.089	-.00	5.20	.3809	-.0071	.0005	.0002	.0014	.0040	0.0000
543	.498	312.855	-.00	6.28	.4593	-.0179	.0000	.0002	.0014	.0040	0.0000
544	.498	313.485	-.00	7.37	.5347	-.0298	.0004	.0004	.0027	.0040	0.0000
545	.498	313.381	-.00	8.48	.6151	-.0312	.0010	.0013	-.0007	.0039	0.0000
547	.497	312.261	-.00	9.54	.6641	-.0220	.0010	.0003	.0007	.0038	0.0000
548	.498	313.638	-.00	10.66	.6916	-.0094	.0005	.0004	-.0001	.0038	0.0000
549	.497	312.504	-.00	11.73	.7255	-.0039	.0003	.0004	-.0009	.0038	0.0000
550	.497	312.084	-.00	12.83	.7586	.0010	.0002	.0003	-.0014	.0037	0.0000
551	.495	309.914	-.00	15.06	.8297	.0082	.0001	.0005	-.0025	.0036	0.0000
552	.497	312.543	-.00	17.16	.8914	.0135	-.0001	.0009	-.0032	.0034	0.0000
553	.499	313.828	-.00	19.27	.9339	.0171	.0003	.0015	-.0041	.0032	0.0000
554	.499	314.421	-.00	21.31	.9884	.0197	.0002	.0016	-.0065	.0025	0.0000
555	.497	311.775	-.00	-.08	-.0220	.0174	.0012	.0006	.0012	.0040	0.0000

TEST= 778 RUN= 41

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	L/D	STAB.PRESS.COEFF CDB	PB-1
537	.497	312.041	-.01	-.05	-.0226	.0169	.0066	.0008	.0007	-1.334	.0040	11.0711
538	.498	313.282	-.00	.98	.0620	.0174	.0132	.0012	.0005	3.570	.0040	11.4757
539	.497	312.222	-.00	2.00	.1393	.0185	.0163	.0010	.0006	7.546	.0040	8.5332
540	.498	313.463	-.00	3.09	.2235	.0206	.0215	.0015	.0005	10.853	.0040	8.1286
541	.498	313.465	-.00	4.12	.2999	.0233	.0272	.0004	.0003	12.852	.0040	7.2091
542	.499	314.089	-.00	5.20	.3800	.0275	.0346	.0005	.0002	13.838	.0039	5.5903
543	.498	312.855	-.00	6.28	.4585	.0325	.0391	.0001	.0002	14.123	.0040	6.4729
544	.498	313.485	-.00	7.37	.5341	.0390	.0473	.0005	.0004	13.702	.0039	6.8959
545	.498	313.381	-.00	8.48	.6130	.0599	.0560	.0012	.0011	10.240	.0039	21.7378
547	.497	312.261	-.00	9.54	.6586	.0885	.0824	.0011	.0001	7.445	.0038	83.4947
548	.498	313.638	-.00	10.66	.6814	.1187	.1234	.0006	.0003	5.741	.0037	82.9430
549	.497	312.504	-.00	11.73	.7111	.1437	.1510	.0004	.0003	4.948	.0037	72.2762
550	.497	312.084	-.00	12.83	.7395	.1695	.1704	.0002	.0003	4.363	.0036	81.6556
551	.495	309.914	-.00	15.06	.7990	.2235	.2011	.0003	.0005	3.576	.0034	89.7476
552	.497	312.543	-.00	17.16	.8478	.2760	.2226	.0001	.0009	3.072	.0033	88.2763
553	.499	313.828	-.00	19.27	.8760	.3243	.2329	.0007	.0014	2.701	.0030	71.9084
554	.499	314.421	-.00	21.31	.9136	.3776	.2317	.0008	.0015	2.420	.0024	79.6326
555	.497	311.775	-.00	-.08	-.0220	.0174	.0054	.0012	.0006	-1.264	.0040	9.9309

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 42

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
556	.292	119.997	.00	-.07	-.0181	.0187	.0020	.0003	-.0017	.0037	0.0000
557	.293	121.349	-.00	.92	.0575	.0184	.0015	.0005	-.0019	.0038	0.0000
558	.292	120.673	.00	1.93	.1247	.0159	.0015	.0003	-.0022	.0037	0.0000
559	.293	121.445	-.00	2.94	.1996	.0111	.0014	.0005	-.0027	.0038	0.0000
560	.291	119.516	-.00	3.95	.2795	.0049	.0014	.0005	.0028	.0037	0.0000
561	.293	121.448	-.00	5.00	.3512	-.0023	.0017	.0003	.0019	.0038	0.0000
562	.290	118.940	.00	6.03	.4161	-.0117	.0002	-.0000	-.0030	.0036	0.0000
563	.291	119.521	-.00	7.07	.4891	-.0232	.0006	.0003	.0021	.0037	0.0000
564	.291	119.524	-.00	8.11	.5546	-.0353	.0001	.0001	.0020	.0037	0.0000
565	.293	120.785	.00	9.15	.6233	-.0481	-.0002	.0006	-.0043	.0037	0.0000
566	.293	121.483	.00	10.33	.6760	-.0525	.0004	-.0013	.0009	.0035	0.0000
567	.289	118.334	.00	11.40	.7488	-.0484	-.0018	-.0019	.0021	.0033	0.0000
568	.290	119.056	-.00	12.51	.8283	-.0439	-.0012	-.0007	.0062	.0033	0.0000
569	.291	119.837	.00	14.71	.8119	-.0050	.0004	.0000	-.0030	.0032	0.0000
570	.297	124.469	-.00	16.74	.8990	.0041	-.0005	.0005	.0019	.0030	0.0000
571	.292	120.382	.00	18.76	.9475	.0092	-.0005	.0005	-.0044	.0028	0.0000
572	.292	120.463	.00	20.81	1.0067	.0130	-.0006	.0007	-.0049	.0024	0.0000

TEST= 778 RUN= 42

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	L/D	STAB.PRESS.COEFF CDB,	PB-1
556	.292	119.997	.00	-.07	-.0180	.0187	.0045	.0020	.0003	-.965	.0037	2.3170
557	.293	121.349	-.00	.92	.0572	.0193	.0123	.0015	.0005	2.957	.0038	1.8573
558	.292	120.673	.00	1.93	.1241	.0201	.0159	.0015	.0003	6.170	.0037	2.1147
559	.293	121.445	-.00	2.94	.1988	.0213	.0203	.0014	.0005	9.321	.0038	2.1331
560	.291	119.516	-.00	3.95	.2785	.0242	.0252	.0014	.0004	11.522	.0037	1.9676
561	.293	121.448	-.00	5.00	.3501	.0283	.0327	.0018	.0001	12.363	.0038	1.4711
562	.290	118.940	.00	6.03	.4151	.0321	.0358	.0002	-.0001	12.949	.0036	1.8941
563	.291	119.521	-.00	7.07	.4882	.0372	.0434	.0006	.0002	13.111	.0037	1.7837
564	.291	119.524	-.00	8.11	.5540	.0433	.0501	.0001	.0001	12.793	.0037	1.9860
565	.293	120.785	.00	9.15	.6230	.0517	.0573	-.0001	.0006	12.058	.0036	3.4387
566	.293	121.483	.00	10.33	.6745	.0696	.0592	.0002	-.0013	9.695	.0035	13.5354
567	.289	118.334	.00	11.40	.7436	.1005	.0802	-.0022	-.0015	7.396	.0033	22.6571
568	.290	119.056	-.00	12.51	.8181	.1365	.1128	-.0013	-.0004	5.993	.0032	27.9540
569	.291	119.837	.00	14.71	.7865	.2013	.1787	.0004	-.0001	3.907	.0031	43.7701
570	.297	124.469	-.00	16.74	.8597	.2629	.2172	-.0003	.0006	3.270	.0029	42.5747
571	.292	120.382	.00	18.76	.8942	.3133	.2358	-.0003	.0006	2.854	.0027	33.1954
572	.292	120.463	.00	20.81	.9364	.3699	.2446	-.0003	.0008	2.532	.0023	32.7356

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 43

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
601	.930	739.454	-.01	-.18	-.0110	.0195	.0007	.0005	.0021	.0061	0.0000
602	.932	740.848	-.01	1.06	.1318	.0203	.0005	.0004	.0019	.0062	0.0000
603	.931	740.444	-.01	2.26	.2528	.0202	.0005	.0001	.0025	.0061	0.0000
604	.932	740.783	-.00	3.52	.3625	.0199	.0009	-.0000	.0026	.0061	0.0000
605	.933	741.973	-.01	4.69	.4589	.0205	.0011	-.0001	.0038	.0060	0.0000
606	.937	744.539	-.01	5.88	.5506	.0210	.0008	-.0003	.0045	.0060	0.0000
607	.935	743.524	-.00	7.12	.6573	.0211	.0009	-.0005	.0041	.0060	0.0000
608	.936	744.295	-.00	8.33	.7442	.0214	-.0002	-.0005	.0035	.0059	0.0000
609	.937	745.042	-.00	9.51	.8163	.0218	-.0017	-.0003	.0025	.0058	0.0000
610	.938	745.513	-.00	10.70	.8924	.0217	-.0029	-.0003	.0021	.0056	0.0000
611	.940	747.417	-.00	11.94	1.0000	.0221	-.0060	-.0003	.0021	.0053	0.0000
612	.929	738.290	-.01	-.11	-.0190	.0185	.0005	.0005	.0013	.0061	0.0000

TEST= 778 RUN= 43

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
601	.930	739.454	-.01	-.18	-.0110	.0195	.0111	.0007	.0005	-.563	.0061	11.5296
602	.932	740.848	-.01	1.06	.1314	.0227	-.0027	.0005	.0004	5.785	.0062	14.2807
603	.931	740.444	-.01	2.26	.2518	.0302	-.0088	.0005	.0001	8.337	.0061	25.4473
604	.932	740.783	-.00	3.52	.3606	.0421	-.0072	.0008	-.0001	8.571	.0061	28.1358
605	.933	741.973	-.01	4.69	.4557	.0579	-.0036	.0011	-.0002	7.865	.0060	26.3226
606	.937	744.539	-.01	5.88	.5455	.0772	-.0016	.0008	-.0004	7.063	.0060	41.2662
607	.935	743.524	-.00	7.12	.6496	.1024	-.0049	.0008	-.0006	6.345	.0059	74.1541
608	.936	744.295	-.00	8.33	.7332	.1290	-.0025	-.0002	-.0004	5.686	.0058	116.5452
609	.937	745.042	-.00	9.51	.8015	.1564	.0044	-.0018	-.0000	5.124	.0057	153.5596
610	.938	745.513	-.00	10.70	.8728	.1869	.0051	-.0029	.0002	4.670	.0055	179.3195
611	.940	747.417	-.00	11.94	.9738	.2286	-.0177	-.0059	.0009	4.260	.0051	226.9633
612	.929	738.290	-.01	-.11	-.0189	.0185	.0125	.0005	.0005	-1.024	.0061	11.5796

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 44

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
626	.864	685.292	-.00	-.17	-.0185	.0168	.0006	.0005	.0003	.0053	0.0000
627	.864	685.288	-.00	.99	.0962	.0155	.0002	.0005	.0001	.0053	0.0000
628	.864	685.297	-.00	2.21	.2252	.0113	.0008	.0006	-.0000	.0053	0.0000
629	.864	685.374	-.00	3.50	.3644	.0056	.0013	.0006	.0001	.0053	0.0000
630	.862	683.974	-.00	4.71	.4979	.0025	.0014	.0003	.0006	.0052	0.0000
631	.862	683.836	-.00	5.94	.6009	.0017	.0007	.0003	.0006	.0051	0.0000
632	.864	685.564	.00	7.13	.6701	.0057	-.0008	-.0003	.0010	.0051	0.0000
633	.866	686.945	-.00	8.29	.7182	.0082	.0021	-.0000	.0017	.0051	0.0000
634	.865	685.845	-.01	9.38	.7298	.0118	.0046	.0001	.0027	.0050	0.0000
635	.864	685.828	.00	10.41	.7340	.0172	-.0031	-.0006	.0018	.0050	0.0000
636	.866	686.915	.00	11.54	.7628	.0196	-.0029	-.0007	.0022	.0049	0.0000
637	.866	687.274	.00	12.64	.7981	.0213	-.0039	-.0006	.0018	.0049	0.0000
638	.866	687.529	.00	13.85	.8384	.0219	-.0043	-.0005	.0017	.0048	0.0000
639	.861	682.845	-.01	-.08	-.0188	.0164	.0003	.0006	.0006	.0052	0.0000

TEST= 778 RUN= 44

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
626	.864	685.292	-.00	-.17	-.0185	.0168	.0100	.0006	.0005	-1.098	.0053	9.7789
627	.864	685.288	-.00	.99	.0960	.0172	.0125	.0002	.0005	5.584	.0053	10.3791
628	.864	685.297	-.00	2.21	.2246	.0199	.0136	.0008	.0006	11.266	.0053	14.8809
629	.864	685.374	-.00	3.50	.3634	.0278	.0079	.0013	.0005	13.069	.0053	20.8830
630	.862	683.974	-.00	4.71	.4960	.0433	.0009	.0015	.0002	11.448	.0052	22.9463
631	.862	683.836	-.00	5.94	.5975	.0638	.0057	.0008	.0002	9.360	.0051	54.2713
632	.864	685.564	.00	7.13	.6642	.0888	.0292	-.0009	-.0002	7.479	.0051	62.2744
633	.866	686.945	-.00	8.29	.7095	.1117	.0550	.0021	-.0003	6.354	.0051	88.1595
634	.865	685.845	-.01	9.38	.7181	.1305	.0850	.0046	-.0007	5.502	.0049	119.0462
635	.864	685.828	.00	10.41	.7188	.1496	.1066	-.0032	-.0000	4.806	.0049	121.0469
636	.866	686.915	.00	11.54	.7435	.1718	.1169	-.0030	-.0001	4.327	.0048	111.5433
637	.866	687.274	.00	12.64	.7740	.1955	.1243	-.0040	.0003	3.959	.0048	110.5429
638	.866	687.529	.00	13.85	.8087	.2219	.1330	-.0043	.0005	3.644	.0047	113.0438
639	.861	682.845	-.01	-.08	-.0188	.0164	.0106	.0003	.0006	-1.146	.0052	10.3541

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 45

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
653	.805	631.881	-.01	-.17	-.0184	.0165	.0005	.0007	.0005	.0048	0.0000
654	.805	631.881	-.01	1.00	.0879	.0155	.0004	.0006	.0011	.0049	0.0000
655	.807	633.958	-.01	2.14	.1978	.0115	.0010	.0007	.0001	.0049	0.0000
656	.806	633.471	-.01	3.34	.3095	.0049	.0010	.0007	.0005	.0049	0.0000
657	.806	633.189	-.01	4.55	.4269	-.0024	.0010	.0007	-.0001	.0049	0.0000
658	.806	633.505	-.00	5.76	.5437	-.0075	.0006	.0004	.0007	.0048	0.0000
659	.807	633.902	.00	6.95	.6206	-.0054	-.0005	-.0002	.0008	.0047	0.0000
660	.807	634.275	-.00	8.08	.6699	-.0002	-.0003	-.0001	.0018	.0047	0.0000
661	.808	634.653	.00	9.17	.6855	.0066	-.0021	-.0004	.0016	.0047	0.0000
662	.808	635.037	.00	10.27	.7185	.0113	-.0027	-.0005	.0015	.0046	0.0000
663	.809	635.928	-.00	11.40	.7493	.0149	-.0030	-.0004	.0024	.0046	0.0000
664	.809	635.807	.00	12.50	.7812	.0175	-.0034	-.0006	.0023	.0045	0.0000
665	.808	635.184	.00	13.65	.8116	.0190	-.0036	-.0005	.0021	.0045	0.0000
666	.809	635.947	-.00	15.91	.8772	.0217	-.0009	.0000	-.0001	.0043	0.0000
667	.810	636.859	-.00	18.08	.9548	.0224	.0005	.0003	.0002	.0040	0.0000
668	.812	638.690	.00	20.24	1.0253	.0223	.0003	.0003	-.0021	.0034	0.0000
669	.815	641.281	.01	22.40	1.1280	.0210	-.0003	.0003	-.0067	.0025	0.0000
670	.806	632.953	-.01	-.08	-.0185	.0163	.0003	.0007	.0007	.0049	0.0000

TEST= 778 RUN= 45

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
653	.805	631.881	-.01	-.17	-.0183	.0165	.0097	.0005	.0007	-1.110	.0048	10.2041
654	.805	631.881	-.01	1.00	.0876	.0170	.0144	.0004	.0006	5.152	.0049	8.7785
655	.807	633.958	-.01	2.14	.1972	.0189	.0168	.0010	.0006	10.446	.0049	9.6539
656	.806	633.471	-.01	3.34	.3087	.0229	.0220	.0010	.0007	13.462	.0049	11.3545
657	.806	633.189	-.01	4.55	.4258	.0315	.0246	.0011	.0007	13.523	.0049	15.7813
658	.806	633.505	-.00	5.76	.5417	.0471	.0272	.0006	.0003	11.503	.0048	24.1343
659	.807	633.902	.00	6.95	.6166	.0698	.0480	-.0005	-.0001	8.835	.0047	54.5214
660	.807	634.275	-.00	8.08	.6633	.0939	.0782	-.0003	-.0001	7.061	.0047	68.7769
661	.808	634.653	.00	9.17	.6757	.1157	.1033	-.0021	-.0001	5.838	.0046	87.5343
662	.808	635.037	.00	10.27	.7050	.1392	.1205	-.0027	-.0000	5.063	.0046	99.2261
663	.809	635.928	-.00	11.40	.7315	.1627	.1335	-.0030	.0002	4.496	.0045	99.9764
664	.809	635.807	.00	12.50	.7589	.1862	.1455	-.0034	.0002	4.076	.0044	92.8486
665	.808	635.184	.00	13.65	.7842	.2100	.1554	-.0036	.0004	3.734	.0044	95.2871
666	.809	635.947	-.00	15.91	.8377	.2613	.1770	-.0008	.0003	3.205	.0041	89.0972
667	.810	636.859	-.00	18.08	.9007	.3177	.1848	.0006	.0001	2.835	.0038	91.1605
668	.812	638.690	.00	20.24	.9543	.3755	.1856	.0004	.0002	2.541	.0032	101.2894
669	.815	641.281	.01	22.40	1.0349	.4492	.1918	-.0002	.0004	2.304	.0023	119.6716
670	.806	632.953	-.01	-.08	-.0185	.0163	.0097	.0003	.0007	-1.132	.0049	9.4038

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 46

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
684	.755	583.780	-.01	-.22	-.0180	.0167	.0003	.0007	.0008	.0047	0.0000
685	.754	582.367	-.01	.88	.0795	.0157	.0002	.0007	.0003	.0046	0.0000
686	.755	583.936	-.01	2.06	.1849	.0119	.0008	.0007	.0004	.0047	0.0000
687	.753	582.009	-.01	3.20	.2853	.0059	.0012	.0008	.0007	.0046	0.0000
688	.753	582.197	-.01	4.37	.3828	-.0014	.0012	.0007	.0001	.0046	0.0000
689	.753	581.856	-.01	5.55	.4867	-.0066	.0014	.0009	.0005	.0046	0.0000
690	.754	582.609	-.00	6.70	.5655	-.0043	.0018	.0004	.0005	.0046	0.0000
691	.754	583.443	.00	7.84	.6115	.0033	-.0007	-.0004	.0017	.0045	0.0000
692	.754	583.198	.00	8.94	.6594	.0075	.0005	-.0005	.0011	.0045	0.0000
693	.754	583.464	.00	10.05	.6940	.0103	-.0021	-.0005	.0015	.0044	0.0000
694	.756	584.810	.00	11.20	.7387	.0122	-.0026	-.0004	.0013	.0044	0.0000
695	.755	584.042	-.00	12.29	.7702	.0143	-.0031	-.0005	.0025	.0043	0.0000
696	.754	583.526	.00	13.44	.8009	.0161	-.0038	-.0007	.0027	.0043	0.0000
697	.757	585.509	-.00	15.68	.8601	.0195	-.0016	.0001	.0011	.0042	0.0000
698	.756	584.475	-.00	17.84	.9167	.0209	.0007	.0001	.0009	.0040	0.0000
699	.759	588.025	.00	19.07	.9635	.0217	.0006	-.0002	.0007	.0037	0.0000
700	.759	587.841	.00	22.06	1.0738	.0204	-.0005	.0003	-.0038	.0026	0.0000
701	.753	582.169	-.01	6.75	.5640	-.0045	.0020	.0005	.0016	.0046	0.0000

TEST= 778 RUN= 46

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
684	.755	583.780	-.01	-.22	-.0180	.0167	.0093	.0003	.0007	-1.074	.0047	8.3533
685	.754	582.367	-.01	.88	.0792	.0170	.0138	.0002	.0007	4.671	.0046	7.8031
686	.755	583.936	-.01	2.06	.1844	.0186	.0184	.0009	.0006	9.932	.0047	7.8531
687	.753	582.009	-.01	3.20	.2845	.0218	.0239	.0012	.0007	13.025	.0046	8.8535
688	.753	582.197	-.01	4.37	.3818	.0278	.0298	.0012	.0006	13.732	.0046	9.3787
689	.753	581.856	-.01	5.55	.4851	.0405	.0357	.0015	.0007	11.975	.0046	13.3303
690	.754	582.609	-.00	6.70	.5621	.0617	.0529	.0019	.0002	9.118	.0046	38.7649
691	.754	583.443	.00	7.84	.6054	.0866	.0827	-.0007	-.0003	6.988	.0045	66.1509
692	.754	583.198	.00	8.94	.6502	.1099	.1028	.0005	-.0006	5.917	.0045	70.6527
693	.754	583.464	.00	10.05	.6816	.1312	.1221	-.0021	-.0001	5.196	.0044	81.4069
694	.756	584.810	.00	11.20	.7222	.1554	.1389	-.0026	.0001	4.648	.0043	91.1607
695	.755	584.042	-.00	12.29	.7495	.1780	.1523	-.0031	.0002	4.211	.0042	89.0972
696	.754	583.526	.00	13.44	.7752	.2018	.1636	-.0039	.0002	3.842	.0042	86.4712
697	.757	585.509	-.00	15.68	.8229	.2512	.1840	-.0015	.0005	3.276	.0040	93.5989
698	.756	584.475	-.00	17.84	.8662	.3006	.1954	.0007	-.0001	2.881	.0038	77.0925
699	.759	588.025	.00	19.07	.9036	.3352	.1957	.0005	-.0004	2.696	.0035	83.8451
700	.759	587.841	.00	22.06	.9875	.4223	.1995	-.0003	.0005	2.338	.0024	101.6646
701	.753	582.169	-.01	6.75	.5606	.0619	.0527	.0021	.0003	9.060	.0045	40.5781

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 47

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
716	.702	529.764	-.00	-.11	-.0173	.0166	.0004	.0006	-.0005	.0045	0.0000
717	.702	529.845	-.00	.94	.0703	.0157	.0004	.0007	-.0010	.0045	0.0000
718	.701	529.462	-.00	2.04	.1689	.0123	.0004	.0006	-.0003	.0045	0.0000
719	.702	530.560	-.00	3.16	.2651	.0065	.0009	.0007	-.0013	.0045	0.0000
720	.701	529.014	-.00	4.28	.3576	-.0009	.0004	.0006	-.0002	.0045	0.0000
721	.703	530.881	-.00	5.43	.4527	-.0071	.0008	.0008	-.0011	.0045	0.0000
722	.702	529.956	-.00	6.58	.5413	-.0060	.0017	.0005	-.0013	.0044	0.0000
723	.702	530.279	.00	7.72	.5884	.0016	-.0009	-.0003	-.0001	.0044	0.0000
724	.702	530.332	.00	8.82	.6421	.0051	-.0003	-.0003	.0013	.0043	0.0000
725	.702	530.706	.00	9.91	.6869	.0074	-.0000	-.0005	.0003	.0043	0.0000
726	.703	531.001	.00	11.03	.7165	.0097	-.0026	-.0006	.0006	.0042	0.0000
727	.704	532.252	.00	12.13	.7550	.0118	-.0033	-.0006	.0007	.0042	0.0000
728	.703	531.011	.00	13.28	.7902	.0136	-.0035	-.0007	.0005	.0041	0.0000
729	.704	531.912	.00	15.54	.8539	.0171	-.0025	-.0000	.0000	.0040	0.0000
730	.705	533.351	-.00	17.67	.9053	.0194	.0004	.0004	-.0006	.0039	0.0000
731	.705	533.273	-.00	19.72	.9696	.0210	.0006	.0003	.0004	.0033	0.0000
732	.705	533.759	.00	21.81	1.0301	.0201	-.0002	.0003	-.0020	.0026	0.0000
733	.702	530.594	-.00	6.60	.5331	-.0050	.0002	-.0000	.0014	.0044	0.0000
734	.701	529.581	-.00	6.60	.5362	-.0051	.0001	.0001	.0015	.0044	0.0000

TEST= 778 RUN= 47

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
716	.702	529.764	-.00	-.11	-.0173	.0166	.0090	.0004	.0006	-1.042	.0045	6.9778
717	.702	529.845	-.00	.94	.0700	.0169	.0133	.0004	.0007	4.146	.0045	7.3779
718	.701	529.462	-.00	2.04	.1683	.0183	.0188	.0004	.0006	9.201	.0045	7.8781
719	.702	530.560	-.00	3.16	.2644	.0211	.0233	.0009	.0006	12.533	.0044	7.4530
720	.701	529.014	-.00	4.28	.3567	.0258	.0304	.0005	.0006	13.815	.0045	7.1779
721	.703	530.881	-.00	5.43	.4513	.0358	.0379	.0008	.0007	12.619	.0044	9.0286
722	.702	529.956	-.00	6.58	.5385	.0560	.0500	.0017	.0003	9.610	.0044	30.0740
723	.702	530.279	.00	7.72	.5828	.0806	.0816	-.0009	-.0002	7.233	.0043	53.5210
724	.702	530.332	.00	8.82	.6337	.1035	.1031	-.0004	-.0003	6.125	.0043	65.4006
725	.702	530.706	.00	9.91	.6754	.1255	.1219	-.0001	-.0005	5.382	.0042	72.0282
726	.703	531.001	.00	11.03	.7015	.1466	.1406	-.0027	-.0001	4.786	.0041	76.2799
727	.704	532.252	.00	12.13	.7357	.1702	.1548	-.0033	.0001	4.323	.0041	81.0318
728	.703	531.011	.00	13.28	.7659	.1947	.1666	-.0036	.0001	3.933	.0040	87.6593
729	.704	531.912	.00	15.54	.8181	.2453	.1876	-.0024	.0006	3.335	.0039	95.0995
730	.705	533.351	-.00	17.67	.8567	.2932	.2031	.0005	.0003	2.922	.0037	73.9038
731	.705	533.273	-.00	19.72	.9056	.3470	.1991	.0006	.0001	2.610	.0031	80.4688
732	.705	533.759	.00	21.81	.9489	.4014	.2013	-.0001	.0003	2.364	.0025	90.9729
733	.702	530.594	-.00	6.60	.5302	.0563	.0522	.0002	-.0001	9.419	.0044	41.0157
734	.701	529.581	-.00	6.60	.5332	.0566	.0518	.0001	.0001	9.422	.0044	40.2029

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 48

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
748	.292	120.670	-.00	.05	-.0171	.0170	.0000	.0008	-.0001	.0038	0.0000
749	.292	120.671	-.00	1.06	.0526	.0168	.0002	.0006	-.0008	.0038	0.0000
750	.293	121.445	-.00	2.02	.1320	.0149	.0019	.0008	-.0027	.0038	0.0000
751	.294	122.121	-.00	3.03	.2087	.0101	.0007	.0006	-.0022	.0038	0.0000
752	.292	120.772	-.00	4.01	.2810	.0051	.0018	.0008	-.0037	.0038	0.0000
753	.293	120.872	.00	5.03	.3685	-.0021	.0006	.0007	-.0033	.0038	0.0000
754	.293	120.874	.00	6.04	.4374	-.0116	-.0001	.0005	-.0032	.0038	0.0000
755	.292	120.691	.00	7.08	.5155	-.0184	.0005	-.0006	-.0041	.0037	0.0000
756	.292	120.722	.00	8.13	.5925	-.0134	.0034	.0012	-.0073	.0037	0.0000
757	.292	120.752	.00	9.16	.6507	-.0102	.0059	.0017	-.0102	.0035	0.0000
758	.292	120.789	.00	10.26	.7028	-.0043	-.0009	-.0008	-.0055	.0036	0.0000
759	.293	120.816	.00	11.28	.7388	-.0036	-.0024	-.0006	-.0050	.0034	0.0000
760	.292	120.075	.00	12.36	.7803	-.0021	-.0031	-.0005	-.0050	.0033	0.0000
761	.294	122.078	.00	14.52	.8367	.0039	-.0000	.0000	-.0090	.0031	0.0000
762	.292	120.221	.00	16.59	.9188	.0080	-.0001	.0003	-.0102	.0029	0.0000
763	.293	120.875	.00	18.62	.9742	.0105	-.0001	.0000	-.0108	.0028	0.0000
764	.294	122.308	.00	20.62	1.0251	.0132	-.0001	.0004	-.0112	.0024	0.0000
765	.293	121.205	.00	8.07	.5906	-.0126	.0033	.0011	-.0131	.0037	0.0000

TEST= 778 RUN= 48

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
748	.292	120.670	-.00	.05	-.0172	.0170	.0050	.0000	.0008	-1.010	.0038	1.4631
749	.292	120.671	-.00	1.06	.0523	.0177	.0127	.0002	.0006	2.950	.0038	1.5381
750	.293	121.445	-.00	2.02	.1313	.0195	.0181	.0019	.0007	6.733	.0038	1.8507
751	.294	122.121	-.00	3.03	.2078	.0211	.0232	.0008	.0006	9.844	.0038	1.6882
752	.292	120.772	-.00	4.01	.2800	.0248	.0313	.0018	.0007	11.296	.0038	1.5881
753	.293	120.872	.00	5.03	.3673	.0303	.0376	.0007	.0006	12.141	.0038	1.5131
754	.293	120.874	.00	6.04	.4361	.0345	.0452	-.0000	.0005	12.632	.0038	1.7757
755	.292	120.691	.00	7.08	.5138	.0453	.0499	.0005	-.0007	11.338	.0036	3.0762
756	.292	120.722	.00	8.13	.5884	.0705	.0613	.0035	.0007	8.346	.0036	14.5558
757	.292	120.752	.00	9.16	.6440	.0936	.0775	.0061	.0008	6.881	.0035	22.4340
758	.292	120.789	.00	10.26	.6924	.1209	.1135	-.0010	-.0006	5.727	.0035	32.3874
759	.293	120.816	.00	11.28	.7252	.1410	.1341	-.0025	-.0001	5.144	.0034	33.5754
760	.292	120.075	.00	12.36	.7626	.1649	.1489	-.0031	.0002	4.624	.0032	37.8895
761	.294	122.078	.00	14.52	.8090	.2136	.1914	-.0000	.0000	3.788	.0030	38.5148
762	.292	120.221	.00	16.59	.8783	.2700	.2227	.0000	.0003	3.253	.0028	34.7008
763	.293	120.875	.00	18.62	.9198	.3210	.2382	-.0001	.0001	2.865	.0027	29.6364
764	.294	122.308	.00	20.62	.9547	.3733	.2435	-.0000	.0004	2.557	.0022	33.3878
765	.293	121.205	.00	8.07	.5866	.0705	.0612	.0035	.0006	8.324	.0036	14.5933

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 49

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
766	.497	312.046	.00	-.17	-.0221	.0180	-.0005	.0007	-.0043	.0040	0.0000
767	.497	312.046	-.00	.86	.0602	.0173	-.0002	.0006	-.0028	.0040	0.0000
768	.497	312.136	.00	1.89	.1430	.0146	.0003	.0005	-.0037	.0040	0.0000
769	.497	312.138	-.00	2.97	.2253	.0097	.0006	.0007	-.0022	.0040	0.0000
770	.497	312.670	-.00	4.03	.3034	.0029	.0006	.0006	-.0027	.0040	0.0000
771	.497	312.675	.00	5.11	.3853	-.0059	.0001	.0003	-.0028	.0040	0.0000
772	.497	312.696	.00	6.20	.4673	-.0136	.0003	.0004	-.0036	.0039	0.0000
773	.498	313.488	-.00	7.31	.5442	-.0106	.0024	.0008	-.0037	.0039	0.0000
774	.497	312.251	-.00	8.40	.6184	-.0073	.0062	.0014	-.0056	.0039	0.0000
775	.498	312.961	-.00	9.49	.6733	-.0054	.0088	.0017	-.0062	.0038	0.0000
776	.497	312.436	-.00	10.58	.7263	-.0038	.0100	.0016	-.0057	.0037	0.0000
777	.497	312.536	.00	11.65	.7386	.0023	-.0029	-.0007	-.0026	.0037	0.0000
778	.498	313.263	.00	12.76	.7792	.0047	-.0037	-.0007	-.0024	.0036	0.0000
779	.498	312.959	.00	14.99	.8540	.0095	-.0024	.0001	-.0048	.0035	0.0000
780	.498	313.699	.00	17.08	.9012	.0129	-.0002	.0004	-.0031	.0034	0.0000
781	.499	313.838	.00	19.14	.9613	.0149	.0002	.0003	-.0040	.0031	0.0000
782	.500	315.400	.00	21.15	1.0053	.0169	-.0013	.0006	-.0027	.0024	0.0000

TEST= 778 RUN= 49

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
766	.497	312.046	.00	-.17	-.0220	.0180	.0069	-.0005	.0007	-1.222	.0040	4.4268
767	.497	312.046	-.00	.86	.0600	.0182	.0134	-.0002	.0006	3.294	.0040	4.2017
768	.497	312.136	.00	1.89	.1424	.0193	.0172	.0003	.0005	7.378	.0040	4.5268
769	.497	312.138	-.00	2.97	.2245	.0214	.0237	.0007	.0006	10.481	.0040	4.5393
770	.497	312.670	-.00	4.03	.3024	.0242	.0298	.0006	.0005	12.485	.0040	4.2267
771	.497	312.675	.00	5.11	.3843	.0284	.0349	.0001	.0003	13.523	.0040	4.4018
772	.497	312.696	.00	6.20	.4660	.0369	.0440	.0003	.0004	12.631	.0039	4.7644
773	.498	313.488	-.00	7.31	.5411	.0587	.0550	.0024	.0005	9.214	.0039	19.6325
774	.497	312.251	-.00	8.40	.6128	.0830	.0709	.0064	.0005	7.380	.0039	36.7016
775	.498	312.961	-.00	9.49	.6650	.1057	.0873	.0089	.0002	6.293	.0037	47.2685
776	.497	312.436	-.00	10.58	.7147	.1297	.1036	.0102	-.0002	5.511	.0036	55.7719
777	.497	312.536	.00	11.65	.7229	.1513	.1485	-.0030	-.0001	4.777	.0036	64.1501
778	.498	313.263	.00	12.76	.7590	.1766	.1620	-.0038	.0001	4.296	.0035	72.9036
779	.498	312.959	.00	14.99	.8225	.2301	.1942	-.0023	.0008	3.574	.0034	93.4114
780	.498	313.699	.00	17.08	.8576	.2771	.2190	-.0000	.0005	3.095	.0032	62.6494
781	.499	313.838	.00	19.14	.9032	.3292	.2279	.0003	.0002	2.744	.0029	62.0867
782	.500	315.400	.00	21.15	.9315	.3785	.2222	-.0010	.0010	2.461	.0023	68.4641

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 50

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
810	.933	741.242	-.00	-.11	.0026	.0204	.0003	.0008	-.0004	.0062	0.0000
811	.930	738.828	-.01	1.13	.1181	.0198	.0006	.0008	-.0002	.0062	0.0000
812	.932	740.224	-.01	2.35	.2266	.0202	.0004	.0006	.0005	.0062	0.0000
813	.933	741.112	-.01	3.59	.3302	.0201	.0002	.0006	.0011	.0061	0.0000
814	.932	740.325	-.01	4.76	.4283	.0196	-.0000	.0005	.0007	.0060	0.0000
815	.935	742.149	-.00	5.97	.5277	.0200	-.0005	.0007	-.0004	.0061	0.0000
816	.935	742.696	-.01	7.21	.6097	.0207	.0030	.0007	.0003	.0060	0.0000
817	.935	742.810	-.01	8.38	.6623	.0212	.0073	.0008	.0001	.0060	0.0000
818	.936	743.438	-.00	9.54	.7204	.0213	.0046	.0006	-.0004	.0059	0.0000
819	.939	745.555	-.00	10.75	.8161	.0215	.0044	.0003	.0004	.0058	0.0000
820	.935	742.470	-.00	11.94	.8590	.0212	.0037	.0002	.0004	.0057	0.0000
821	.926	735.421	-.01	.06	-.0009	.0175	-.0001	.0008	.0009	.0062	0.0000

TEST= 778 RUN= 50

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-L
810	.933	741.242	-.00	-.11	.0026	.0204	.0037	.0003	.0008	.129	.0062	12.1015
811	.930	738.828	-.01	1.13	.1177	.0221	-.0037	.0006	.0008	5.324	.0062	12.1791
812	.932	740.224	-.01	2.35	.2256	.0295	-.0113	.0004	.0006	7.648	.0062	19.8977
813	.933	741.112	-.01	3.59	.3283	.0407	-.0175	.0002	.0006	8.069	.0061	24.0479
814	.932	740.325	-.01	4.76	.4252	.0551	-.0217	.0000	.0005	7.713	.0060	35.7816
815	.935	742.149	-.00	5.97	.5227	.0748	-.0266	-.0004	.0007	6.992	.0060	72.1453
816	.935	742.696	-.01	7.21	.6023	.0970	-.0262	.0031	.0003	6.207	.0059	99.4907
817	.935	742.810	-.01	8.38	.6521	.1175	-.0110	.0074	-.0002	5.552	.0059	111.9028
818	.936	743.438	-.00	9.54	.7069	.1404	-.0066	.0046	-.0001	5.035	.0059	133.0422
819	.939	745.555	-.00	10.75	.7978	.1734	-.0181	.0044	-.0005	4.602	.0057	166.3997
820	.935	742.470	-.00	11.94	.8360	.1985	-.0101	.0037	-.0006	4.212	.0056	188.1208
821	.926	735.421	-.01	.06	-.0009	.0175	.0045	-.0001	.0008	-.052	.0062	11.1319

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 51

					BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
836	.863	683.779	-.00	-.16	-.0088	.0159	.0006	.0008	-.0006	.0052	0.0000
837	.864	684.915	-.00	1.02	.0897	.0150	.0003	.0008	-.0012	.0053	0.0000
838	.864	684.918	-.00	2.17	.1917	.0120	.0002	.0007	-.0008	.0053	0.0000
839	.863	683.982	-.00	3.39	.2995	.0073	.0007	.0008	-.0009	.0052	0.0000
840	.861	682.392	-.00	4.61	.4114	.0033	.0006	.0007	-.0015	.0052	0.0000
841	.862	683.309	-.00	5.83	.5259	.0026	.0016	.0009	-.0017	.0051	0.0000
842	.863	683.743	-.00	7.01	.6011	.0037	.0019	.0007	-.0005	.0051	0.0000
843	.864	684.825	-.00	8.10	.6303	.0082	-.0001	.0003	-.0009	.0051	0.0000
844	.864	685.181	-.00	9.23	.6725	.0111	.0022	.0006	-.0011	.0051	0.0000
845	.865	685.492	-.00	10.32	.6942	.0158	.0002	.0003	-.0011	.0051	0.0000
846	.867	687.213	-.00	11.42	.7253	.0191	-.0005	.0003	-.0010	.0051	0.0000
847	.865	685.959	.00	12.53	.7644	.0204	-.0012	.0002	-.0008	.0050	0.0000
848	.865	685.938	.00	13.71	.8096	.0220	-.0006	.0002	-.0007	.0049	0.0000
849	.863	683.529	-.00	-.20	-.0124	.0158	-.0000	.0008	-.0011	.0053	0.0000

TEST= 778 RUN= 51

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
836	.863	683.779	-.00	-.16	-.0087	.0159	.0052	.0006	.0008	-.549	.0052	13.0712
837	.864	684.915	-.00	1.02	.0894	.0166	.0089	.0003	.0008	5.381	.0053	13.4203
838	.864	684.918	-.00	2.17	.1911	.0192	.0118	.0002	.0007	9.952	.0053	13.9633
839	.863	683.982	-.00	3.39	.2986	.0250	.0128	.0008	.0007	11.938	.0052	14.6227
840	.861	682.392	-.00	4.61	.4098	.0363	.0099	.0007	.0007	11.277	.0052	25.0176
841	.862	683.309	-.00	5.83	.5229	.0560	.0021	.0017	.0007	9.340	.0051	40.2421
842	.863	683.743	-.00	7.01	.5961	.0771	.0077	.0020	.0005	7.732	.0051	67.2968
843	.864	684.825	-.00	8.10	.6228	.0969	.0241	-.0001	.0003	6.424	.0051	147.4909
844	.864	685.181	-.00	9.23	.6620	.1188	.0348	.0023	.0002	5.572	.0050	159.7090
845	.865	685.492	-.00	10.32	.6802	.1399	.0435	.0003	.0003	4.862	.0050	175.1272
846	.867	687.213	-.00	11.42	.7071	.1623	.0484	-.0004	.0004	4.357	.0050	191.6117
847	.865	685.959	.00	12.53	.7418	.1857	.0532	-.0011	.0005	3.994	.0049	175.7087
848	.865	685.938	.00	13.71	.7813	.2132	.0558	-.0005	.0003	3.664	.0048	150.8845
849	.863	683.529	-.00	-.20	-.0124	.0159	.0046	-.0000	.0008	-.779	.0053	12.7609

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 52

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
870	.805	632.715	-.00	-.24	-.0061	.0159	.0000	.0008	-.0002	.0048	0.0000
871	.805	632.865	-.01	.91	.0827	.0150	.0001	.0008	.0002	.0048	0.0000
872	.807	634.361	-.01	2.09	.1766	.0118	.0001	.0008	.0005	.0048	0.0000
873	.804	631.444	-.01	3.24	.2667	.0069	.0004	.0007	.0006	.0048	0.0000
874	.806	633.125	-.01	4.43	.3621	.0006	.0005	.0007	.0009	.0048	0.0000
875	.807	634.368	-.01	5.62	.4645	-.0049	.0002	.0007	.0014	.0048	0.0000
876	.807	634.255	-.00	6.83	.5575	-.0049	.0009	.0006	.0002	.0047	0.0000
877	.809	635.995	-.00	7.99	.6186	-.0014	.0003	.0003	.0011	.0047	0.0000
878	.808	635.842	-.00	9.08	.6613	.0037	-.0002	.0002	.0020	.0047	0.0000
879	.807	634.613	-.00	10.20	.7023	.0077	.0001	.0002	.0012	.0047	0.0000
880	.808	635.496	-.00	11.32	.7312	.0125	-.0007	.0001	.0025	.0046	0.0000
881	.808	634.994	-.00	12.41	.7602	.0153	-.0009	-.0000	.0023	.0046	0.0000
882	.809	636.818	-.00	13.57	.7940	.0172	.0000	.0000	.0022	.0045	0.0000
883	.809	636.348	-.00	15.81	.8568	.0199	.0004	.0001	.0023	.0044	0.0000
884	.813	639.704	-.01	18.01	.9326	.0222	.0003	-.0001	.0042	.0040	0.0000
885	.811	638.025	-.01	20.14	1.0117	.0237	.0008	-.0002	.0057	.0031	0.0000
886	.813	639.917	-.01	22.27	1.0940	.0230	.0017	-.0011	.0110	.0023	0.0000
887	.807	634.276	-.01	6.93	.5610	-.0048	.0010	.0007	.0012	.0047	0.0000

TEST= 778 RUN= 52

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
870	.805	632.715	-.00	-.24	-.0060	.0159	.0052	.0000	.0008	-.377	.0048	8.8820
871	.805	632.865	-.01	.91	.0825	.0163	.0099	.0001	.0008	5.055	.0048	9.4444
872	.807	634.361	-.01	2.09	.1761	.0183	.0147	.0001	.0008	9.646	.0048	9.6771
873	.804	631.444	-.01	3.24	.2659	.0219	.0194	.0005	.0007	12.116	.0048	10.5886
874	.806	633.125	-.01	4.43	.3610	.0285	.0243	.0006	.0007	12.653	.0048	11.6552
875	.807	634.368	-.01	5.62	.4627	.0406	.0266	.0003	.0007	11.384	.0048	15.7475
876	.807	634.255	-.00	6.83	.5541	.0614	.0287	.0009	.0005	9.021	.0047	47.1269
877	.809	635.995	-.00	7.99	.6128	.0846	.0369	.0003	.0002	7.244	.0046	61.7693
878	.808	635.842	-.00	9.08	.6525	.1080	.0458	-.0002	.0002	6.039	.0046	86.1089
879	.807	634.613	-.00	10.20	.6898	.1319	.0537	.0001	.0002	5.229	.0046	106.6665
880	.808	635.496	-.00	11.32	.7145	.1558	.0617	-.0007	.0002	4.587	.0046	121.6000
881	.808	634.994	-.00	12.41	.7392	.1783	.0682	-.0009	.0002	4.146	.0045	132.0727
882	.809	636.818	-.00	13.57	.7678	.2031	.0746	.0000	-.0000	3.781	.0044	123.9272
883	.809	636.348	-.00	15.81	.8189	.2526	.0834	.0004	.0000	3.242	.0042	125.0909
884	.813	639.704	-.01	18.01	.8800	.3095	.0799	.0003	-.0002	2.843	.0038	151.8545
885	.811	638.025	-.01	20.14	.9416	.3707	.0766	.0007	-.0005	2.540	.0029	171.0545
886	.813	639.917	-.01	22.27	1.0036	.4359	.0794	.0012	-.0017	2.302	.0021	175.9999
887	.807	634.276	-.01	6.93	.5575	.0629	.0293	.0010	.0006	8.862	.0047	47.9027

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 53

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
901	.753	582.255	-.00	-.12	-.0065	.0156	.0000	.0008	-.0001	.0045	0.0000
902	.753	582.334	-.01	1.00	.0742	.0148	.0002	.0009	.0002	.0046	0.0000
903	.754	583.381	-.00	2.10	.1646	.0119	.0002	.0009	-.0005	.0046	0.0000
904	.753	582.422	-.00	3.26	.2476	.0074	.0002	.0009	-.0001	.0046	0.0000
905	.755	584.085	-.00	4.39	.3335	.0012	.0006	.0008	.0001	.0046	0.0000
906	.753	582.728	-.00	5.55	.4250	-.0047	.0005	.0008	-.0006	.0046	0.0000
907	.754	583.049	-.01	6.74	.5137	-.0053	.0015	.0013	-.0007	.0045	0.0000
908	.753	582.234	-.00	7.85	.5733	-.0013	.0010	.0006	.0002	.0045	0.0000
909	.754	582.938	-.00	8.99	.6269	.0028	.0001	.0003	.0014	.0045	0.0000
910	.754	583.567	-.00	10.10	.6851	.0053	.0003	.0003	.0005	.0044	0.0000
911	.755	583.841	-.00	11.27	.7235	.0077	-.0001	.0002	.0003	.0044	0.0000
912	.755	584.278	-.00	12.34	.7582	.0107	.0004	.0002	.0006	.0043	0.0000
913	.757	586.153	-.00	13.48	.7826	.0137	.0008	.0002	.0011	.0043	0.0000
914	.755	584.483	-.00	15.73	.8388	.0176	.0009	.0001	.0016	.0042	0.0000
915	.757	586.476	-.00	17.87	.9056	.0202	.0007	-.0000	.0025	.0039	0.0000
916	.758	587.102	-.01	20.00	.9718	.0226	.0001	.0002	.0026	.0031	0.0000
917	.759	588.135	-.01	22.10	1.0454	.0221	.0011	-.0006	.0062	.0024	0.0000
918	.752	581.572	-.01	-.10	-.0106	.0155	.0002	.0009	.0010	.0046	0.0000

TEST= 778 RUN= 53

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
901	.753	582.255	-.00	-.12	-.0065	.0156	.0045	.0000	.0008	-.417	.0045	14.8942
902	.753	582.334	-.01	1.00	.0740	.0161	.0098	.0003	.0009	4.599	.0046	14.3512
903	.754	583.381	-.00	2.10	.1641	.0180	.0147	.0002	.0009	9.131	.0046	14.7003
904	.753	582.422	-.00	3.26	.2468	.0215	.0201	.0003	.0009	11.491	.0046	15.3209
905	.755	584.085	-.00	4.39	.3325	.0267	.0255	.0007	.0007	12.441	.0046	16.6784
906	.753	582.728	-.00	5.55	.4234	.0364	.0293	.0006	.0007	11.617	.0045	23.2722
907	.754	583.049	-.01	6.74	.5108	.0551	.0313	.0016	.0012	9.279	.0045	38.0118
908	.753	582.234	-.00	7.85	.5681	.0771	.0391	.0010	.0005	7.371	.0044	58.5693
909	.754	582.938	-.00	8.99	.6188	.1007	.0483	.0002	.0003	6.144	.0045	77.7695
910	.754	583.567	-.00	10.10	.6736	.1254	.0571	.0004	.0003	5.373	.0044	96.9695
911	.755	583.841	-.00	11.27	.7081	.1490	.0666	-.0001	.0002	4.753	.0043	105.8907
912	.755	584.278	-.00	12.34	.7383	.1725	.0754	.0004	.0001	4.281	.0042	107.8301
913	.757	586.153	-.00	13.48	.7578	.1958	.0832	.0009	.0000	3.871	.0042	104.1452
914	.755	584.483	-.00	15.73	.8026	.2443	.0914	.0009	-.0001	3.285	.0040	112.0967
915	.757	586.476	-.00	17.87	.8557	.2971	.0920	.0006	-.0003	2.880	.0037	122.4727
916	.758	587.102	-.01	20.00	.9054	.3537	.0854	.0002	.0002	2.560	.0030	138.4727
917	.759	588.135	-.01	22.10	.9602	.4138	.0887	.0008	-.0009	2.320	.0022	149.2363
918	.752	581.572	-.01	-.10	-.0105	.0155	.0044	.0002	.0009	-.680	.0046	14.7391

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 54

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
932	.702	530.488	-.01	-.07	-.0095	.0158	-.0001	.0010	-.0000	.0044	0.0000
933	.702	530.023	-.01	1.01	.0704	.0150	-.0003	.0010	.0007	.0044	0.0000
934	.703	531.281	-.00	2.12	.1505	.0122	.0001	.0008	-.0003	.0044	0.0000
935	.701	529.723	-.00	3.20	.2311	.0078	.0005	.0009	.0001	.0044	0.0000
936	.701	529.818	-.00	4.34	.3136	.0019	.0008	.0009	-.0009	.0044	0.0000
937	.701	529.461	-.00	5.44	.3917	-.0043	.0007	.0008	-.0002	.0043	0.0000
938	.702	530.205	-.00	6.63	.4867	-.0061	.0007	.0008	-.0010	.0043	0.0000
939	.703	531.956	-.00	7.75	.5581	-.0023	.0004	.0005	-.0000	.0043	0.0000
940	.702	530.585	-.00	8.87	.6091	.0010	.0008	.0005	.0003	.0043	0.0000
941	.702	530.783	-.00	9.96	.6581	.0037	.0004	.0005	-.0001	.0042	0.0000
942	.702	530.451	-.00	11.11	.7075	.0059	-.0003	.0003	-.0002	.0042	0.0000
943	.702	530.199	-.00	12.20	.7403	.0088	.0004	.0003	.0002	.0042	0.0000
944	.703	531.537	-.00	13.36	.7747	.0112	.0009	.0001	.0006	.0041	0.0000
945	.703	531.464	-.00	15.60	.8320	.0154	.0013	.0002	.0010	.0040	0.0000
946	.703	531.788	-.00	17.72	.8913	.0178	.0014	-.0001	.0017	.0038	0.0000
947	.704	532.725	-.00	19.81	.9550	.0211	.0004	.0003	.0009	.0031	0.0000
948	.706	534.854	-.00	21.88	1.0157	.0215	-.0002	.0002	.0022	.0024	0.0000
949	.700	528.077	-.00	-.10	-.0114	.0158	.0003	.0010	-.0003	.0043	0.0000

TEST= 778 RUN= 54

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
932	.702	530.488	-.01	-.07	-.0094	.0158	.0042	-.0001	.0010	-.598	.0044	7.1172
933	.702	530.023	-.01	1.01	.0701	.0162	.0105	-.0003	.0010	4.317	.0044	6.2639
934	.703	531.281	-.00	2.12	.1499	.0178	.0154	.0001	.0008	8.427	.0044	6.8845
935	.701	529.723	-.00	3.20	.2303	.0207	.0211	.0005	.0009	11.149	.0044	6.4385
936	.701	529.818	-.00	4.34	.3126	.0256	.0268	.0008	.0008	12.192	.0044	7.0009
937	.701	529.461	-.00	5.44	.3904	.0328	.0324	.0008	.0008	11.897	.0043	7.5051
938	.702	530.205	-.00	6.63	.4841	.0501	.0339	.0008	.0007	9.661	.0043	26.1233
939	.703	531.956	-.00	7.75	.5533	.0730	.0398	.0005	.0005	7.576	.0042	55.9511
940	.702	530.585	-.00	8.87	.6017	.0949	.0499	.0009	.0004	6.338	.0042	74.0847
941	.702	530.783	-.00	9.96	.6475	.1175	.0599	.0005	.0004	5.511	.0042	91.3453
942	.702	530.451	-.00	11.11	.6931	.1422	.0706	-.0002	.0004	4.875	.0041	105.5028
943	.702	530.199	-.00	12.20	.7217	.1650	.0783	.0005	.0002	4.373	.0041	105.6968
944	.703	531.537	-.00	13.36	.7511	.1898	.0871	.0009	-.0001	3.957	.0040	106.6665
945	.703	531.464	-.00	15.60	.7972	.2386	.0965	.0013	-.0001	3.342	.0039	112.0967
946	.703	531.788	-.00	17.72	.8436	.2883	.0996	.0013	-.0005	2.926	.0036	112.2907
947	.704	532.725	-.00	19.81	.8913	.3435	.0926	.0004	.0001	2.595	.0029	123.3452
948	.706	534.854	-.00	21.88	.9327	.3977	.0931	-.0002	.0002	2.345	.0022	144.2909
949	.700	528.077	-.00	-.10	-.0114	.0159	.0035	.0003	.0010	-.717	.0043	6.5160

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 55

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
964	.292	120.479	-.00	-.09	-.0095	.0168	.0005	.0005	-.0004	.0037	0.0000
965	.293	121.252	-.00	.91	.0479	.0166	.0018	.0007	-.0019	.0036	0.0000
966	.293	121.155	-.00	1.86	.1134	.0147	.0005	.0004	-.0016	.0037	0.0000
967	.292	120.577	.00	2.90	.1811	.0116	.0012	.0002	-.0028	.0037	0.0000
968	.293	121.351	.00	3.88	.2462	.0074	.0017	.0004	-.0038	.0036	0.0000
969	.293	121.353	.00	4.89	.3125	.0015	.0016	.0001	-.0044	.0036	0.0000
970	.293	121.355	.00	5.96	.3777	-.0056	.0003	.0004	-.0040	.0036	0.0000
971	.293	121.362	.00	7.00	.4449	-.0117	.0010	-.0004	-.0050	.0036	0.0000
972	.293	121.280	.00	8.04	.5199	-.0149	.0002	.0005	-.0051	.0037	0.0000
973	.293	121.308	.00	9.09	.5961	-.0116	.0009	.0002	-.0062	.0035	0.0000
974	.293	121.332	.00	10.18	.6527	-.0114	.0014	.0004	-.0073	.0035	0.0000
975	.293	121.358	.00	11.21	.7095	-.0117	.0019	.0001	-.0083	.0034	0.0000
976	.293	121.486	.00	12.33	.7555	-.0096	.0021	.0003	-.0091	.0034	0.0000
977	.292	120.206	.00	14.49	.8351	-.0011	.0013	-.0004	-.0038	.0032	0.0000
978	.292	120.860	.00	16.56	.9114	.0042	-.0001	-.0000	-.0038	.0031	0.0000
979	.292	120.250	.00	18.58	.9602	.0070	-.0001	-.0004	-.0044	.0029	0.0000
980	.293	121.007	.00	20.58	1.0241	.0113	-.0002	-.0006	-.0048	.0025	0.0000
981	.293	121.378	.00	8.01	.5195	-.0139	.0002	.0005	-.0051	.0037	0.0000

TEST= 778 RUN= 55

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
964	.292	120.479	-.00	-.09	-.0095	.0168	.0024	.0005	.0005	-.566	.0037	.0388
965	.293	121.252	-.00	.91	.0476	.0173	.0089	.0019	.0006	2.749	.0036	-.0776
966	.293	121.155	-.00	1.86	.1128	.0184	.0156	.0005	.0004	6.128	.0036	.0776
967	.292	120.577	.00	2.90	.1802	.0207	.0225	.0012	.0001	8.702	.0037	0.0000
968	.293	121.351	.00	3.88	.2451	.0241	.0291	.0017	.0003	10.185	.0036	.5818
969	.293	121.353	.00	4.89	.3112	.0282	.0330	.0017	.0000	11.038	.0036	.7951
970	.293	121.355	.00	5.96	.3762	.0337	.0369	.0003	.0004	11.165	.0036	.9309
971	.293	121.362	.00	7.00	.4430	.0426	.0380	.0010	-.0005	10.390	.0036	2.6568
972	.293	121.280	.00	8.04	.5169	.0580	.0422	.0002	.0004	8.913	.0036	12.7221
973	.293	121.308	.00	9.09	.5905	.0827	.0463	.0009	.0001	7.141	.0035	30.7777
974	.293	121.332	.00	10.18	.6444	.1042	.0555	.0015	.0002	6.185	.0035	31.1270
975	.293	121.358	.00	11.21	.6982	.1265	.0648	.0019	-.0002	5.520	.0033	36.6543
976	.293	121.486	.00	12.33	.7401	.1519	.0737	.0021	-.0001	4.871	.0033	45.6724
977	.292	120.206	.00	14.49	.8089	.2079	.0949	.0012	-.0007	3.891	.0031	55.1754
978	.292	120.860	.00	16.56	.8724	.2638	.1094	-.0001	.0000	3.307	.0029	54.0117
979	.292	120.250	.00	18.58	.9079	.3127	.1186	-.0002	-.0003	2.904	.0028	47.7087
980	.293	121.007	.00	20.58	.9547	.3706	.1167	-.0004	-.0005	2.576	.0024	53.0420
981	.293	121.378	.00	8.01	.5164	.0587	.0422	.0002	.0004	8.798	.0036	14.3900

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 56

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
982	.497	312.326	-.00	-.10	-.0150	.0173	.0000	.0007	-.0023	.0039	0.0000
983	.496	311.706	-.00	.89	.0547	.0167	.0002	.0008	-.0008	.0039	0.0000
984	.498	313.125	-.00	1.92	.1349	.0145	-.0001	.0007	-.0014	.0040	0.0000
985	.497	312.507	-.00	3.05	.1976	.0105	-.0001	.0006	-.0019	.0039	0.0000
986	.497	312.510	.00	4.09	.2636	.0056	-.0002	.0006	-.0026	.0039	0.0000
987	.498	313.132	-.00	5.12	.3362	-.0017	-.0002	.0005	-.0009	.0039	0.0000
988	.496	311.991	.00	6.19	.4078	-.0091	.0002	.0002	-.0018	.0039	0.0000
989	.497	312.750	-.00	7.34	.4874	-.0112	.0002	.0001	-.0001	.0039	0.0000
990	.498	313.450	-.00	8.45	.5630	-.0084	.0002	.0004	-.0009	.0038	0.0000
991	.498	313.709	.00	9.50	.6247	-.0061	.0005	.0002	-.0018	.0038	0.0000
992	.498	313.714	.00	9.53	.6320	-.0060	.0005	.0003	-.0019	.0038	0.0000
993	.497	312.555	-.00	10.67	.6703	-.0042	.0003	.0002	-.0001	.0037	0.0000
994	.497	312.733	-.00	11.69	.7173	-.0019	.0002	.0002	-.0007	.0037	0.0000
995	.498	313.454	.00	12.77	.7591	.0011	.0004	.0002	-.0014	.0037	0.0000
996	.498	313.141	.00	15.01	.8424	.0064	.0010	.0001	-.0006	.0036	0.0000
997	.498	313.172	.00	17.19	.8958	.0099	.0011	.0001	-.0014	.0035	0.0000
998	.499	314.550	-.00	19.22	.9552	.0134	.0004	.0000	.0009	.0031	0.0000
999	.499	314.638	-.00	21.27	1.0403	.0164	.0003	-.0001	.0004	.0025	0.0000
0	.498	313.186	-.00	8.40	.5672	-.0086	.0001	.0004	-.0008	.0038	0.0000

TEST= 778 RUN= 56

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
982	.497	312.326	-.00	-.10	-.0150	.0173	.0028	.0000	.0007	-.865	.0039	3.1223
983	.496	311.706	-.00	.89	.0544	.0176	.0087	.0002	.0008	3.094	.0039	3.3356
984	.498	313.125	-.00	1.92	.1344	.0190	.0147	-.0001	.0007	7.071	.0040	3.1998
985	.497	312.507	-.00	3.05	.1968	.0210	.0204	-.0001	.0006	9.352	.0039	3.0835
986	.497	312.510	.00	4.09	.2625	.0244	.0261	-.0001	.0007	10.765	.0039	3.1223
987	.498	313.132	-.00	5.12	.3350	.0283	.0309	-.0002	.0006	11.843	.0038	3.1611
988	.496	311.991	.00	6.19	.4064	.0350	.0357	.0002	.0002	11.622	.0039	3.7235
989	.497	312.750	-.00	7.34	.4849	.0512	.0384	.0002	.0001	9.477	.0038	16.6396
990	.498	313.450	-.00	8.45	.5582	.0744	.0443	.0003	.0004	7.499	.0038	37.7013
991	.498	313.709	.00	9.50	.6172	.0970	.0542	.0005	.0002	6.359	.0037	62.8359
992	.498	313.714	.00	9.53	.6243	.0988	.0544	.0005	.0002	6.320	.0037	62.6423
993	.497	312.555	-.00	10.67	.6595	.1199	.0660	.0003	.0002	5.500	.0036	73.5029
994	.497	312.733	-.00	11.69	.7028	.1435	.0756	.0003	.0002	4.898	.0036	83.9756
995	.498	313.454	.00	12.77	.7400	.1689	.0838	.0004	.0001	4.381	.0036	91.1513
996	.498	313.141	.00	15.01	.8120	.2244	.1006	.0010	-.0001	3.619	.0035	94.0604
997	.498	313.172	.00	17.19	.8528	.2742	.1102	.0011	-.0002	3.111	.0033	85.5271
998	.499	314.550	-.00	19.22	.8976	.3271	.1121	.0004	-.0001	2.744	.0030	93.4786
999	.499	314.638	-.00	21.27	.9634	.3927	.1098	.0003	-.0002	2.453	.0023	94.4483
0	.498	313.186	-.00	8.40	.5623	.0744	.0434	.0002	.0004	7.562	.0038	39.9512

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 57

					BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
74	.932	740.028	-.01	-.26	-.0003	.0205	.0005	.0007	.0003	.0061	0.0000
75	.930	738.756	-.01	1.01	.1332	.0201	.0012	.0007	.0002	.0061	0.0000
76	.931	739.669	-.00	2.22	.2574	.0209	.0022	.0006	-.0002	.0061	0.0000
77	.931	739.557	-.00	3.45	.3756	.0211	.0013	.0005	.0003	.0061	0.0000
78	.932	739.996	-.00	4.62	.4697	.0216	.0002	.0005	-.0003	.0060	0.0000
79	.934	741.817	-.00	5.86	.5733	.0220	.0010	.0004	.0005	.0060	0.0000
80	.936	743.129	-.00	7.10	.6554	.0242	.0009	.0003	-.0001	.0060	0.0000
81	.933	740.754	-.00	8.24	.7059	.0242	-.0043	-.0000	.0011	.0059	0.0000
82	.938	744.655	-.00	9.44	.7879	.0256	-.0007	.0001	.0001	.0059	0.0000
83	.937	744.351	-.00	10.58	.8488	.0258	.0005	.0001	.0003	.0057	0.0000
84	.929	737.336	-.01	-.20	-.0048	.0182	.0011	.0008	.0007	.0061	0.0000

TEST= 778 RUN= 57

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
74	.932	740.028	-.01	-.26	-.0002	.0205	.0056	.0005	.0007	-.012	.0061	15.3230
75	.930	738.756	-.01	1.01	.1328	.0224	-.0044	.0012	.0006	5.928	.0061	18.7999
76	.931	739.669	-.00	2.22	.2564	.0309	-.0164	.0023	.0006	8.309	.0061	16.4306
77	.931	739.557	-.00	3.45	.3736	.0437	-.0238	.0013	.0005	8.557	.0061	21.6614
78	.932	739.996	-.00	4.62	.4664	.0594	-.0231	.0003	.0005	7.855	.0060	36.3077
79	.934	741.817	-.00	5.86	.5680	.0804	-.0272	.0010	.0003	7.064	.0060	62.6153
80	.936	743.129	-.00	7.10	.6474	.1050	-.0201	.0009	.0002	6.165	.0060	88.7691
81	.933	740.754	-.00	8.24	.6951	.1251	.0036	-.0043	.0006	5.557	.0059	116.4614
82	.938	744.655	-.00	9.44	.7730	.1544	-.0004	-.0007	.0002	5.005	.0058	140.9229
83	.937	744.351	-.00	10.58	.8296	.1812	.0062	.0006	-.0000	4.579	.0057	183.3844
84	.929	737.336	-.01	-.20	-.0047	.0182	.0068	.0011	.0008	-.261	.0061	16.6153

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 58

						BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
PGINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
98	.861	682.149	-.01	-.27	-.0091	.0164	.0003	.0008	.0005	.0052	0.0000
99	.862	682.796	-.00	.92	.1008	.0154	.0006	.0007	-.0005	.0052	0.0000
100	.865	685.065	-.00	2.12	.2134	.0116	.0010	.0006	-.0005	.0053	0.0000
101	.862	682.934	-.01	3.36	.3406	.0066	.0010	.0007	-.0001	.0052	0.0000
102	.863	683.294	-.01	4.59	.4655	.0039	.0008	.0008	-.0004	.0051	0.0000
103	.862	682.424	-.00	5.80	.5708	.0051	.0008	.0004	-.0000	.0051	0.0000
104	.862	682.927	-.00	6.96	.6418	.0077	.0002	.0003	-.0004	.0051	0.0000
105	.864	684.718	-.00	8.11	.6649	.0128	-.0009	.0002	-.0002	.0051	0.0000
106	.865	685.004	-.00	9.17	.6854	.0171	-.0006	.0003	-.0001	.0051	0.0000
107	.865	685.394	-.00	10.29	.7186	.0199	.0009	.0004	-.0001	.0051	0.0000
108	.867	687.045	-.00	11.41	.7462	.0225	.0008	.0004	.0007	.0050	0.0000
109	.866	686.397	-.00	12.52	.7829	.0242	.0003	.0003	.0008	.0050	0.0000
110	.867	686.721	-.00	13.70	.8289	.0249	.0003	.0004	.0003	.0049	0.0000
111	.867	686.679	-.00	16.03	.9152	.0258	.0004	.0003	.0005	.0047	0.0000
112	.870	689.118	-.00	18.23	1.0048	.0266	.0004	.0007	-.0001	.0043	0.0000
113	.863	683.508	-.01	-.19	-.0156	.0158	.0007	.0008	.0013	.0052	0.0000

TEST= 778 RUN= 58

						STABILITY AXIS COEFFICIENTS				STAB.PRESS.COEFF		
PGINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
98	.861	682.149	-.01	-.27	-.0090	.0164	.0088	.0003	.0008	-.551	.0052	12.5230
99	.862	682.796	-.00	.92	.1005	.0170	.0117	.0006	.0007	5.910	.0052	12.9845
100	.865	685.065	-.00	2.12	.2128	.0194	.0143	.0010	.0006	10.953	.0053	18.2768
101	.862	682.934	-.01	3.36	.3396	.0265	.0129	.0010	.0007	12.810	.0052	20.0614
102	.863	683.294	-.01	4.59	.4637	.0412	.0053	.0008	.0008	11.255	.0051	24.0769
103	.862	682.424	-.00	5.80	.5674	.0628	.0090	.0008	.0004	9.037	.0051	59.6922
104	.862	682.927	-.00	6.96	.6361	.0854	.0249	.0003	.0003	7.445	.0050	83.5384
105	.864	684.718	-.00	8.11	.6564	.1064	.0501	-.0008	.0003	6.167	.0050	115.8460
106	.865	685.004	-.00	9.17	.6739	.1261	.0691	-.0006	.0004	5.342	.0050	116.0768
107	.865	685.394	-.00	10.29	.7035	.1479	.0818	.0010	.0002	4.758	.0050	113.0768
108	.867	687.045	-.00	11.41	.7270	.1697	.0895	.0008	.0002	4.285	.0049	120.9230
109	.866	686.397	-.00	12.52	.7591	.1933	.0964	.0003	.0003	3.926	.0049	129.6922
110	.867	686.721	-.00	13.70	.7995	.2206	.1027	.0004	.0003	3.625	.0047	131.5383
111	.867	686.679	-.00	16.03	.8725	.2774	.1133	.0005	.0002	3.145	.0045	109.1537
112	.870	689.118	-.00	18.23	.9461	.3396	.1125	.0006	.0005	2.786	.0040	125.9999
113	.863	683.508	-.01	-.19	-.0156	.0159	.0083	.0007	.0008	-.980	.0052	15.8460

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 59

					BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
127	.805	631.189	-.01	-.25	-.0114	.0160	.0005	.0008	.0005	.0047	0.0000
128	.807	632.611	-.01	.91	.0887	.0152	.0008	.0008	.0006	.0048	0.0000
129	.806	631.613	-.01	2.05	.1910	.0118	.0008	.0008	-.0001	.0048	0.0000
130	.806	631.778	-.01	3.24	.2950	.0060	.0012	.0008	-.0001	.0048	0.0000
131	.806	631.824	-.00	4.41	.4004	-.0005	.0008	.0007	-.0004	.0047	0.0000
132	.805	630.972	-.00	5.61	.5089	-.0044	.0011	.0006	-.0002	.0047	0.0000
133	.806	632.417	-.00	6.81	.5928	-.0026	.0013	.0006	-.0002	.0047	0.0000
134	.806	632.193	-.00	7.93	.6385	.0031	.0004	.0005	-.0002	.0047	0.0000
135	.807	632.915	-.00	9.05	.6790	.0085	.0011	.0006	-.0004	.0047	0.0000
136	.807	633.346	-.00	10.13	.7116	.0126	.0001	.0004	-.0001	.0047	0.0000
137	.807	632.708	-.00	11.25	.7408	.0164	.0005	.0004	.0002	.0046	0.0000
138	.808	634.015	-.00	12.35	.7711	.0194	.0004	.0003	.0010	.0046	0.0000
139	.810	635.645	-.00	13.51	.8012	.0211	.0007	.0004	.0002	.0046	0.0000
140	.810	635.388	-.00	15.79	.8726	.0236	.0004	.0007	-.0014	.0044	0.0000
141	.806	631.984	-.00	17.94	.9437	.0241	.0003	.0009	-.0006	.0040	0.0000
142	.811	636.932	-.00	20.06	1.0283	.0253	-.0001	.0011	-.0019	.0033	0.0000
143	.807	632.877	-.01	-.24	-.0115	.0158	.0004	.0008	.0006	.0048	0.0000
144	.806	631.379	-.01	-.25	-.0130	.0158	.0008	.0007	.0002	.0047	0.0000

TEST= 778 RUN= 59

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
127	.805	631.189	-.01	-.25	-.0113	.0160	.0076	.0005	.0008	-.708	.0047	13.3845
128	.807	632.611	-.01	.91	.0885	.0166	.0125	.0008	.0008	5.319	.0048	11.5999
129	.806	631.613	-.01	2.05	.1905	.0186	.0164	.0009	.0008	10.256	.0048	13.7230
130	.806	631.778	-.01	3.24	.2942	.0227	.0216	.0013	.0007	12.954	.0048	11.1691
131	.806	631.824	-.00	4.41	.3992	.0304	.0263	.0008	.0007	13.153	.0047	13.3230
132	.805	630.972	-.00	5.61	.5069	.0454	.0293	.0011	.0005	11.170	.0047	21.6614
133	.806	632.417	-.00	6.81	.5889	.0678	.0415	.0014	.0004	8.688	.0047	46.1538
134	.806	632.193	-.00	7.93	.6320	.0911	.0625	.0005	.0005	6.937	.0046	69.2307
135	.807	632.915	-.00	9.05	.6692	.1152	.0789	.0011	.0004	5.811	.0046	81.3845
136	.807	633.346	-.00	10.13	.6983	.1376	.0924	.0002	.0003	5.076	.0046	92.9230
137	.807	632.708	-.00	11.25	.7233	.1606	.1044	.0006	.0003	4.503	.0045	97.8461
138	.808	634.015	-.00	12.35	.7491	.1839	.1135	.0004	.0003	4.073	.0045	104.1537
139	.810	635.645	-.00	13.51	.7741	.2077	.1224	.0007	.0002	3.727	.0045	100.9230
140	.810	635.388	-.00	15.79	.8333	.2601	.1324	.0006	.0006	3.203	.0042	93.2307
141	.806	631.984	-.00	17.94	.8904	.3136	.1363	.0006	.0008	2.839	.0038	106.1537
142	.811	636.932	-.00	20.06	.9572	.3765	.1342	.0003	.0010	2.542	.0031	118.4614
143	.807	632.877	-.01	-.24	-.0114	.0159	.0082	.0004	.0008	-.722	.0048	10.8615
144	.806	631.379	-.01	-.25	-.0129	.0159	.0075	.0008	.0008	-.815	.0047	11.5076

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 60

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
158	.753	580.533	-.01	-.22	-.0145	.0159	.0004	.0008	.0007	.0046	0.0000
159	.753	581.214	-.01	.88	.0745	.0152	.0003	.0008	.0001	.0046	0.0000
160	.752	580.248	-.01	2.02	.1740	.0120	.0007	.0008	.0002	.0046	0.0000
161	.753	580.781	-.00	3.18	.2652	.0069	.0009	.0008	-.0006	.0046	0.0000
162	.754	581.487	-.00	4.33	.3617	.0002	.0008	.0007	.0000	.0046	0.0000
163	.753	581.139	-.00	5.49	.4571	-.0040	.0010	.0008	-.0009	.0045	0.0000
164	.753	580.897	-.00	6.67	.5383	-.0028	.0018	.0010	-.0012	.0045	0.0000
165	.754	582.107	-.00	7.81	.6037	-.0001	.0023	.0010	-.0013	.0045	0.0000
166	.756	583.388	-.00	8.93	.6397	.0077	.0008	.0005	.0005	.0045	0.0000
167	.755	582.603	-.00	10.05	.6870	.0100	.0006	.0005	.0001	.0045	0.0000
168	.755	582.887	-.00	11.17	.7290	.0126	.0005	.0006	.0007	.0044	0.0000
169	.755	583.160	-.00	12.23	.7575	.0154	.0006	.0006	.0001	.0044	0.0000
170	.756	583.993	-.00	13.41	.7872	.0176	.0005	.0004	-.0002	.0043	0.0000
172	.757	584.529	-.00	15.64	.8482	.0208	.0003	.0006	-.0007	.0042	0.0000
173	.758	585.547	-.00	17.78	.9146	.0222	.0004	.0008	-.0013	.0040	0.0000
174	.756	584.023	-.00	19.90	.9956	.0243	.0002	.0009	-.0015	.0032	0.0000
175	.755	583.088	-.00	22.02	1.0681	.0229	.0003	.0009	-.0009	.0025	0.0000
176	.753	580.768	-.01	-.22	-.0129	.0158	-.0001	.0009	.0011	.0046	0.0000

TEST= 778 RUN= 60

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
158	.753	580.533	-.01	-.22	-.0144	.0159	.0075	.0004	.0008	-.905	.0046	11.9999
159	.753	581.214	-.01	.88	.0743	.0163	.0134	.0003	.0008	4.551	.0046	10.0307
160	.752	580.248	-.01	2.02	.1735	.0181	.0181	.0008	.0008	9.563	.0046	11.8768
161	.753	580.781	-.00	3.18	.2644	.0216	.0229	.0010	.0007	12.233	.0046	12.2153
162	.754	581.487	-.00	4.33	.3606	.0275	.0294	.0009	.0006	13.102	.0045	11.3538
163	.753	581.139	-.00	5.49	.4554	.0398	.0325	.0011	.0007	11.447	.0045	15.9383
164	.753	580.897	-.00	6.67	.5350	.0598	.0445	.0019	.0008	8.948	.0045	37.0769
165	.754	582.107	-.00	7.81	.5981	.0820	.0618	.0024	.0007	7.297	.0045	58.7692
166	.756	583.388	-.00	8.93	.6308	.1068	.0829	.0008	.0004	5.904	.0044	78.6153
167	.755	582.603	-.00	10.05	.6747	.1297	.0964	.0006	.0004	5.202	.0044	86.7691
168	.755	582.887	-.00	11.17	.7128	.1536	.1109	.0006	.0004	4.641	.0043	92.6153
169	.755	583.160	-.00	12.23	.7371	.1755	.1218	.0007	.0004	4.201	.0043	95.6922
170	.756	583.993	-.00	13.41	.7616	.1997	.1320	.0005	.0003	3.815	.0042	91.9999
172	.757	584.529	-.00	15.64	.8112	.2486	.1423	.0005	.0005	3.262	.0041	88.7691
173	.758	585.547	-.00	17.78	.8641	.3004	.1465	.0006	.0007	2.876	.0038	93.9999
174	.756	584.023	-.00	19.90	.9279	.3618	.1429	.0005	.0008	2.565	.0030	103.0768
175	.755	583.088	-.00	22.02	.9816	.4217	.1490	.0006	.0007	2.328	.0023	113.8460
176	.753	580.768	-.01	-.22	-.0128	.0158	.0082	-.0001	.0009	-.811	.0046	11.7538

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 61

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
190	.702	529.017	-.01	-.23	-.0181	.0160	.0080	.0002	.0008	-1.135	.0044	9.0612
191	.701	528.470	-.01	.86	.0686	.0163	.0124	.0001	.0008	4.202	.0044	9.6307
192	.701	528.004	-.00	1.98	.1582	.0177	.0191	.0009	.0006	8.917	.0044	10.0615
193	.702	528.804	-.00	3.13	.2511	.0211	.0247	.0008	.0007	11.913	.0044	10.2153
194	.701	528.351	-.00	4.21	.3379	.0258	.0312	.0004	.0006	13.075	.0044	7.9999
195	.702	528.959	-.00	5.36	.4286	.0361	.0366	.0007	.0005	11.882	.0044	10.2461
196	.701	528.825	-.00	6.54	.5137	.0569	.0452	.0007	.0003	9.031	.0043	32.8461
197	.703	530.046	-.00	7.68	.5695	.0795	.0626	.0002	.0004	7.165	.0043	55.5384
198	.701	528.759	-.00	8.77	.6152	.1008	.0816	.0006	.0003	6.104	.0042	69.2307
199	.702	529.596	-.00	9.87	.6578	.1231	.0981	.0005	.0003	5.345	.0042	78.1538
200	.701	528.162	-.00	11.00	.6938	.1458	.1137	.0004	.0004	4.759	.0042	80.1538
201	.704	531.601	-.00	12.07	.7253	.1685	.1248	.0003	.0005	4.304	.0041	91.8461
202	.704	531.293	-.00	13.25	.7524	.1928	.1341	.0001	.0005	3.903	.0040	93.6922
203	.702	529.572	-.00	15.47	.8034	.2416	.1476	.0003	.0005	3.325	.0039	87.6922
204	.703	530.538	-.00	17.59	.8507	.2912	.1534	.0003	.0007	2.921	.0037	86.3076
205	.704	530.933	-.00	19.69	.9011	.3466	.1491	.0000	.0008	2.600	.0030	98.3076
206	.705	532.378	-.00	21.76	.9520	.4042	.1512	.0002	.0008	2.355	.0023	105.4614
207	.701	528.632	-.01	-.25	-.0137	.0158	.0083	.0003	.0008	-.864	.0043	11.4918
208	.701	528.632	-.00	.83	.0670	.0162	.0130	.0010	.0007	4.139	.0044	9.9692

TEST= 778 RUN= 61

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
190	.702	529.017	-.01	-.23	-.0182	.0159	.0002	.0008	.0010	.0044	0.0000
191	.701	528.470	-.01	.86	.0689	.0153	.0001	.0008	.0004	.0044	0.0000
192	.701	528.004	-.00	1.98	.1587	.0123	.0009	.0006	.0004	.0044	0.0000
193	.702	528.804	-.00	3.13	.2519	.0073	.0008	.0008	-.0002	.0044	0.0000
194	.701	528.351	-.00	4.21	.3389	.0009	.0003	.0006	-.0005	.0044	0.0000
195	.702	528.959	-.00	5.36	.4301	-.0042	.0007	.0005	-.0001	.0044	0.0000
196	.701	528.825	-.00	6.54	.5168	-.0020	.0006	.0004	.0005	.0044	0.0000
197	.703	530.046	-.00	7.68	.5750	.0027	.0002	.0004	.0001	.0043	0.0000
198	.701	528.759	-.00	8.77	.6234	.0057	.0006	.0004	.0003	.0043	0.0000
199	.702	529.596	-.00	9.87	.6692	.0085	.0005	.0003	-.0004	.0043	0.0000
200	.701	528.162	-.00	11.00	7089	.0107	.0003	.0005	.0005	.0042	0.0000
201	.704	531.601	-.00	12.07	.7445	.0131	.0001	.0005	.0000	.0042	0.0000
202	.704	531.293	-.00	13.25	.7765	.0152	-.0000	.0005	-.0002	.0041	0.0000
203	.702	529.572	-.00	15.47	.8387	.0186	.0002	.0006	-.0012	.0041	0.0000
204	.703	530.538	-.00	17.59	.8990	.0204	.0000	.0007	-.0016	.0039	0.0000
205	.704	530.933	-.00	19.69	.9652	.0228	-.0003	.0008	-.0015	.0032	0.0000
206	.705	532.378	-.00	21.76	1.0340	.0225	-.0001	.0008	-.0022	.0025	0.0000
207	.701	528.632	-.01	-.25	-.0137	.0158	.0004	.0008	.0008	.0043	0.0000
208	.701	528.632	-.00	.83	.0672	.0152	.0010	.0007	-.0003	.0044	0.0000

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 62

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
222	.292	120.181	-.00	-.07	-.0195	.0158	.0005	.0004	-.0007	.0038	0.0000
223	.292	120.086	-.00	.89	.0569	.0163	.0005	.0007	-.0012	.0038	0.0000
224	.292	120.086	-.00	1.87	.1242	.0141	.0009	.0009	-.0021	.0038	0.0000
225	.292	120.087	-.00	2.87	.1813	.0110	.0004	.0006	-.0022	.0038	0.0000
226	.292	120.090	.00	3.89	.2678	.0064	.0008	.0004	-.0033	.0038	0.0000
227	.293	120.767	.00	4.90	.3331	-.0003	.0007	.0002	-.0037	.0038	0.0000
228	.292	120.095	.00	5.94	.4018	-.0081	.0007	.0004	-.0042	.0037	0.0000
229	.292	120.109	.00	6.98	.4766	-.0110	.0002	.0002	-.0043	.0037	0.0000
230	.292	120.134	.00	8.06	.5512	-.0087	.0001	.0005	-.0049	.0037	0.0000
231	.293	120.744	.00	9.14	.6319	-.0059	.0001	.0003	-.0059	.0037	0.0000
232	.292	120.092	.00	10.27	.6617	-.0054	.0001	.0004	-.0065	.0036	0.0000
233	.291	119.444	.00	11.30	.7113	-.0044	.0000	.0001	-.0073	.0036	0.0000
234	.291	119.475	.00	12.44	.7474	-.0024	.0000	.0002	-.0078	.0035	0.0000
235	.292	120.220	.00	14.60	.8241	.0019	-.0003	.0007	-.0088	.0033	0.0000
236	.291	119.515	.00	16.69	.8829	.0062	-.0003	.0006	-.0096	.0032	0.0000
237	.292	120.170	.00	18.73	.9516	.0092	-.0001	.0005	-.0107	.0031	0.0000
238	.293	120.936	.00	20.80	1.0311	.0126	-.0002	.0008	-.0111	.0025	0.0000
239	.291	119.219	.00	-.06	-.0095	.0182	.0003	.0004	-.0065	.0038	0.0000

TEST= 778 RUN= 62

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
222	.292	120.181	-.00	-.07	-.0195	.0158	.0110	.0005	.0004	-1.230	.0038	1.2461
223	.292	120.086	-.00	.89	.0567	.0172	.0158	.0005	.0006	3.292	.0038	2.3999
224	.292	120.086	-.00	1.87	.1237	.0181	.0199	.0009	.0008	6.825	.0038	1.8615
225	.292	120.087	-.00	2.87	.1805	.0200	.0265	.0004	.0006	9.006	.0038	2.2614
226	.292	120.090	.00	3.89	.2667	.0246	.0350	.0008	.0004	10.851	.0038	1.7384
227	.293	120.767	.00	4.90	.3319	.0282	.0389	.0008	.0001	11.778	.0038	1.2153
228	.292	120.095	.00	5.94	.4005	.0335	.0432	.0007	.0003	11.960	.0037	1.0307
229	.292	120.109	.00	6.98	.4744	.0470	.0478	.0002	.0002	10.087	.0037	6.1228
230	.292	120.134	.00	8.06	.5469	.0687	.0555	.0002	.0004	7.964	.0037	18.2152
231	.293	120.744	.00	9.14	.6248	.0945	.0760	.0001	.0003	6.608	.0036	27.4615
232	.292	120.092	.00	10.27	.6520	.1127	.0903	.0001	.0004	5.786	.0035	30.3077
233	.291	119.444	.00	11.30	.6984	.1351	.1092	.0000	.0001	5.171	.0035	32.6154
234	.291	119.475	.00	12.44	.7304	.1586	.1206	.0001	.0002	4.605	.0034	37.0769
235	.292	120.220	.00	14.60	.7970	.2096	.1463	-.0001	.0007	3.802	.0032	40.6153
236	.291	119.515	.00	16.69	.8439	.2595	.1656	-.0001	.0006	3.251	.0031	36.6153
237	.292	120.170	.00	18.73	.8983	.3142	.1813	.0001	.0005	2.859	.0029	31.7692
238	.293	120.936	.00	20.80	.9595	.3778	.1817	.0001	.0008	2.539	.0024	32.1538
239	.291	119.219	.00	-.06	-.0095	.0183	.0086	.0003	.0004	-.520	.0038	2.4768

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 64

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
253	.497	311.732	-.00	.05	-.0158	.0155	.0000	.0009	.0021	.0039	0.0000
254	.497	311.733	-.00	1.10	.0623	.0147	.0008	.0008	.0008	.0039	0.0000
255	.497	311.206	-.01	2.16	.1396	.0126	.0006	.0009	.0027	.0039	0.0000
256	.497	312.010	-.00	3.24	.2199	.0083	.0000	.0008	.0025	.0039	0.0000
257	.497	311.924	-.00	4.29	.2864	.0023	.0000	.0007	.0020	.0039	0.0000
258	.497	311.932	-.00	5.37	.3673	-.0054	-.0001	.0004	.0038	.0039	0.0000
259	.497	311.962	-.00	6.51	.4404	-.0106	-.0004	.0005	.0035	.0038	0.0000
260	.497	311.510	-.00	7.58	.5334	-.0083	.0008	.0006	.0017	.0038	0.0000
261	.497	312.218	-.00	8.71	.5902	-.0048	.0014	.0008	.0026	.0038	0.0000
262	.497	311.685	-.00	9.75	.6485	-.0021	.0007	.0007	.0023	.0038	0.0000
263	.497	311.593	-.00	10.92	.6877	-.0008	.0004	.0005	.0018	.0037	0.0000
264	.497	311.685	-.00	11.95	.7306	.0014	.0004	.0006	.0012	.0037	0.0000
265	.498	312.940	-.00	13.10	.7643	.0040	.0004	.0006	.0006	.0036	0.0000
266	.498	312.982	-.00	15.29	.8452	.0083	.0001	.0008	.0020	.0035	0.0000
267	.499	313.824	-.00	17.44	.9031	.0118	-.0002	.0011	.0015	.0034	0.0000
268	.497	312.094	-.00	19.49	.9633	.0144	.0002	.0006	.0007	.0031	0.0000
269	.499	314.209	-.00	21.51	1.0219	.0173	-.0009	.0012	.0016	.0024	0.0000
270	.497	311.642	-.01	.08	-.0122	.0153	.0001	.0009	.0043	.0039	0.0000

TEST= 778 RUN= 64

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
253	.497	311.732	-.00	.05	-.0158	.0155	.0073	.0000	.0009	-1.019	.0039	5.8921
254	.497	311.733	-.00	1.10	.0620	.0159	.0136	.0008	.0007	3.896	.0039	4.7998
255	.497	311.206	-.01	2.16	.1390	.0178	.0187	.0006	.0009	7.806	.0039	5.0767
256	.497	312.010	-.00	3.24	.2191	.0208	.0264	.0001	.0008	10.555	.0039	5.3229
257	.497	311.924	-.00	4.29	.2854	.0237	.0308	.0001	.0007	12.045	.0039	4.5383
258	.497	311.932	-.00	5.37	.3662	.0291	.0373	-.0000	.0004	12.599	.0039	3.9075
259	.497	311.962	-.00	6.51	.4388	.0394	.0433	-.0003	.0006	11.142	.0038	6.4767
260	.497	311.510	-.00	7.58	.5299	.0621	.0506	.0008	.0005	8.529	.0038	26.6154
261	.497	312.218	-.00	8.71	.5841	.0846	.0652	.0015	.0006	6.902	.0037	47.3846
262	.497	311.685	-.00	9.75	.6395	.1077	.0860	.0008	.0006	5.935	.0037	59.9999
263	.497	311.593	-.00	10.92	.6754	.1296	.1055	.0005	.0004	5.213	.0036	58.9230
264	.497	311.685	-.00	11.95	.7145	.1526	.1191	.0005	.0005	4.683	.0036	64.4615
265	.498	312.940	-.00	13.10	.7426	.1769	.1305	.0005	.0005	4.197	.0035	76.3076
266	.498	312.982	-.00	15.29	.8131	.2309	.1526	.0003	.0007	3.522	.0034	80.6153
267	.499	313.824	-.00	17.44	.8580	.2820	.1667	.0002	.0011	3.043	.0033	71.8461
268	.497	312.094	-.00	19.49	.9033	.3350	.1722	.0004	.0005	2.696	.0029	72.3076
269	.499	314.209	-.00	21.51	.9443	.3909	.1654	-.0004	.0014	2.416	.0023	84.7691
270	.497	311.642	-.01	.08	-.0122	.0153	.0063	.0001	.0009	-.796	.0039	5.0921

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 65

					BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
324	.884	701.784	-.01	.20	.2688	.0242	.0016	.0006	.0008	.0055	0.0000
325	.883	700.826	-.01	1.37	.3678	.0232	.0016	.0004	.0012	.0055	0.0000
326	.883	700.575	-.01	2.53	.4567	.0214	.0018	.0003	.0013	.0054	0.0000
327	.883	700.784	-.00	3.71	.5433	.0188	.0015	.0003	.0010	.0055	0.0000
328	.883	700.549	-.00	4.86	.6362	.0151	-.0000	.0002	.0017	.0054	0.0000
329	.884	701.947	-.00	6.03	.7272	.0115	.0022	.0001	.0016	.0053	0.0000
330	.884	701.847	-.00	7.23	.8101	.0068	.0013	.0002	.0017	.0053	0.0000
331	.883	700.534	-.00	8.42	.8782	.0038	.0008	.0001	.0013	.0052	0.0000
332	.883	700.910	-.00	9.56	.9360	.0020	.0005	.0001	.0018	.0052	0.0000
333	.887	703.957	-.00	10.68	.9595	.0040	-.0002	.0001	.0024	.0052	0.0000
334	.886	703.228	-.01	11.81	.9716	.0050	-.0009	.0004	.0010	.0052	0.0000
335	.883	700.463	-.01	.26	.2674	.0231	.0014	.0006	.0009	.0055	0.0000

TEST= 778 RUN= 65

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
324	.884	701.784	-.01	.20	.2687	.0251	-.1235	.0016	.0006	10.704	.0055	16.3999
325	.883	700.826	-.01	1.37	.3672	.0320	-.1193	.0016	.0004	11.466	.0055	29.1538
326	.883	700.575	-.01	2.53	.4553	.0415	-.1078	.0018	.0003	10.976	.0054	31.4615
327	.883	700.784	-.00	3.71	.5410	.0538	-.1000	.0015	.0002	10.047	.0055	38.1538
328	.883	700.549	-.00	4.86	.6326	.0689	-.0963	-.0000	.0002	9.176	.0054	37.0000
329	.884	701.947	-.00	6.03	.7219	.0878	-.0995	.0022	-.0001	8.219	.0053	43.6153
330	.884	701.847	-.00	7.23	.8028	.1087	-.0915	.0013	.0001	7.386	.0053	73.8461
331	.883	700.534	-.00	8.42	.8682	.1324	-.0724	.0008	-.0000	6.559	.0052	114.9230
332	.883	700.910	-.00	9.56	.9227	.1574	-.0535	.0005	.0001	5.864	.0052	238.1536
333	.887	703.957	-.00	10.68	.9422	.1819	-.0190	-.0002	.0001	5.181	.0051	158.7691
334	.886	703.228	-.01	11.81	.9500	.2038	.0163	-.0008	.0006	4.662	.0051	248.7690
335	.883	700.463	-.01	.26	.2673	.0243	-.1210	.0014	.0006	11.009	.0055	15.3230

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 66

					BODY AXIS COEFFICIENTS				BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
386	.823	647.599	-.01	.25	.2432	.0199	.0009	.0007	.0010	.0049	0.0000
387	.822	647.185	-.01	1.42	.3559	.0163	.0013	.0008	.0011	.0049	0.0000
388	.822	646.709	-.01	2.59	.4670	.0102	.0015	.0007	.0016	.0049	0.0000
389	.822	646.770	-.01	3.81	.5924	.0028	.0019	.0006	.0020	.0049	0.0000
390	.824	648.901	-.01	4.97	.6921	-.0024	.0022	.0004	.0022	.0049	0.0000
391	.824	648.986	-.01	6.14	.7763	-.0083	.0037	.0004	.0025	.0049	0.0000
392	.821	646.198	-.01	7.31	.8475	-.0135	.0021	.0002	.0030	.0048	0.0000
393	.823	647.916	-.01	8.45	.9042	-.0134	.0010	.0001	.0031	.0048	0.0000
394	.824	648.814	-.01	9.59	.9431	-.0098	.0021	.0004	.0021	.0048	0.0000
395	.824	648.696	-.01	10.69	.9835	-.0078	.0029	.0008	.0028	.0048	0.0000
396	.825	649.742	-.01	11.82	1.0052	-.0055	.0042	.0004	.0030	.0047	0.0000
397	.826	650.551	-.01	12.85	.9854	-.0021	.0078	.0002	.0045	.0047	0.0000
398	.823	647.291	-.01	.27	.2417	.0195	.0013	.0008	.0017	.0049	0.0000

TEST= 778 RUN= 66

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
386	.823	647.599	-.01	.25	.2431	.0210	-.1053	.0009	.0007	11.592	.0049	13.2307
387	.822	647.185	-.01	1.42	.3554	.0251	-.1045	.0014	.0007	14.156	.0049	15.6614
388	.822	646.709	-.01	2.59	.4660	.0314	-.1051	.0015	.0006	14.862	.0049	18.6152
389	.822	646.770	-.01	3.81	.5909	.0422	-.1098	.0019	.0005	13.999	.0049	17.8768
390	.824	648.901	-.01	4.97	.6897	.0576	-.1059	.0022	.0002	11.972	.0049	34.0000
391	.824	648.986	-.01	6.14	.7727	.0747	-.0960	.0037	-.0000	10.339	.0048	46.7692
392	.821	646.198	-.01	7.31	.8424	.0945	-.0780	.0021	-.0000	8.916	.0048	66.4615
393	.823	647.916	-.01	8.45	.8964	.1196	-.0540	.0010	-.0001	7.498	.0048	212.7690
394	.824	648.814	-.01	9.59	.9315	.1475	-.0195	.0021	.0001	6.314	.0047	185.0767
395	.824	648.696	-.01	10.69	.9679	.1748	.0022	.0030	.0002	5.537	.0047	360.9227
396	.825	649.742	-.01	11.82	.9850	.2005	.0258	.0042	-.0005	4.912	.0046	401.0765
397	.826	650.551	-.01	12.85	.9612	.2172	.0560	.0076	-.0015	4.426	.0046	171.2306
398	.823	647.291	-.01	.27	.2416	.0207	-.1055	.0013	.0008	11.677	.0049	13.1384

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 67

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
412	.769	596.665	-.01	.17	.2231	.0194	.0014	.0007	.0004	.0047	0.0000
413	.768	595.788	-.01	1.29	.3281	.0160	.0018	.0007	.0007	.0047	0.0000
414	.770	597.259	-.01	2.45	.4335	.0097	.0017	.0006	.0014	.0047	0.0000
415	.770	597.258	-.01	3.60	.5396	.0004	.0013	.0005	.0013	.0047	0.0000
416	.769	596.397	-.01	4.77	.6432	-.0112	.0016	.0005	.0018	.0047	0.0000
417	.768	595.809	-.00	5.96	.7547	-.0236	.0020	.0003	.0021	.0046	0.0000
418	.769	596.860	-.00	7.14	.8487	-.0304	.0006	.0002	.0013	.0046	0.0000
419	.771	598.281	-.00	8.28	.8892	-.0287	-.0001	-.0001	.0018	.0046	0.0000
420	.769	597.059	-.01	9.39	.9387	-.0269	.0009	.0005	.0012	.0046	0.0000
421	.771	598.497	-.00	10.51	.9935	-.0226	.0007	.0005	.0006	.0046	0.0000
423	.773	600.733	-.01	11.60	.9954	-.0158	.0132	.0004	.0019	.0046	0.0000
424	.770	596.944	-.01	.15	.2249	.0194	.0013	.0007	.0005	.0047	0.0000

TEST= 778 RUN= 67

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
412	.769	596.665	-.01	.17	.2231	.0201	-.0987	.0014	.0007	11.090	.0047	11.6922
413	.768	595.788	-.01	1.29	.3277	.0235	-.0957	.0018	.0007	13.973	.0047	11.8768
414	.770	597.259	-.01	2.45	.4327	.0282	-.0934	.0018	.0006	15.344	.0047	13.7845
415	.770	597.258	-.01	3.60	.5385	.0342	-.0928	.0013	.0005	15.734	.0047	14.3384
416	.769	596.397	-.01	4.77	.6419	.0423	-.0920	.0016	.0004	15.169	.0047	17.3229
417	.768	595.809	-.00	5.96	.7531	.0550	-.0897	.0020	.0001	13.703	.0046	20.5842
418	.769	596.860	-.00	7.14	.8459	.0753	-.0768	.0006	.0002	11.229	.0046	61.6922
419	.771	598.281	-.00	8.28	.8841	.0997	-.0387	-.0001	-.0001	8.864	.0046	142.6152
420	.769	597.059	-.01	9.39	.9305	.1267	-.0134	.0010	.0003	7.346	.0045	381.2304
421	.771	598.497	-.00	10.51	.9810	.1590	.0034	.0008	.0003	6.171	.0045	485.3841
423	.773	600.733	-.01	11.60	.9783	.1847	.0286	.0130	-.0023	5.297	.0045	209.8459
424	.770	596.944	-.01	.15	.2249	.0200	-.0991	.0013	.0007	11.267	.0047	10.5845

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 68

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
438	.717	544.323	-.01	.10	.2209	.0194	.0020	.0008	.0012	.0044	0.0000
439	.716	543.402	-.01	1.23	.3206	.0160	.0015	.0007	.0010	.0045	0.0000
440	.717	544.482	-.01	2.32	.4154	.0100	.0017	.0007	.0016	.0045	0.0000
441	.718	545.021	-.01	3.49	.5184	.0009	.0012	.0006	.0015	.0045	0.0000
442	.717	543.946	-.01	4.60	.6088	-.0100	.0017	.0007	.0019	.0045	0.0000
443	.717	544.120	-.01	5.75	.7020	-.0233	.0017	.0006	.0013	.0045	0.0000
444	.717	543.716	-.01	6.94	.7980	-.0365	.0019	.0007	.0018	.0045	0.0000
445	.717	544.365	-.00	8.08	.8693	-.0396	-.0003	-.0003	.0028	.0044	0.0000
446	.718	545.222	-.00	9.19	.8927	-.0314	-.0016	-.0002	.0027	.0045	0.0000
447	.718	545.118	-.01	10.27	.9296	-.0281	.0045	.0005	.0017	.0044	0.0000
448	.717	544.359	-.00	11.39	.9561	-.0213	.0055	.0003	.0013	.0044	0.0000
449	.720	546.787	-.00	12.47	.9703	-.0128	.0061	.0000	.0026	.0043	0.0000
450	.718	545.442	-.00	13.58	.9679	-.0073	.0017	.0001	.0033	.0043	0.0000
451	.721	548.151	-.01	15.81	1.0070	-.0004	.0008	.0002	.0032	.0042	0.0000
452	.720	547.380	-.01	17.92	1.0515	.0008	.0013	.0003	.0035	.0039	0.0000
453	.720	547.465	-.00	19.99	1.1026	.0038	.0011	.0003	.0020	.0033	0.0000
454	.722	548.626	-.00	22.09	1.1657	.0050	.0007	.0002	.0020	.0027	0.0000
455	.719	546.482	-.00	8.17	.8703	-.0380	.0002	-.0000	.0036	.0045	0.0000

TEST= 778 RUN= 68

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
438	.717	544.323	-.01	.10	.2209	.0198	-.0941	.0020	.0008	11.166	.0044	8.9227
439	.716	543.402	-.01	1.23	.3202	.0229	-.0902	.0015	.0007	14.005	.0045	9.9381
440	.717	544.482	-.01	2.32	.4147	.0269	-.0879	.0017	.0007	15.440	.0045	10.9230
441	.718	545.021	-.01	3.49	.5174	.0324	-.0848	.0012	.0005	15.945	.0045	11.1384
442	.717	543.946	-.01	4.60	.6077	.0389	-.0820	.0018	.0005	15.640	.0045	11.6922
443	.717	544.120	-.01	5.75	.7008	.0471	-.0762	.0018	.0004	14.863	.0045	13.8768
444	.717	543.716	-.01	6.94	.7965	.0602	-.0703	.0020	.0004	13.237	.0045	15.4460
445	.717	544.365	-.00	8.08	.8663	.0829	-.0469	-.0003	-.0003	10.447	.0044	74.9230
446	.718	545.222	-.00	9.19	.8863	.1115	-.0046	-.0016	.0001	7.948	.0044	168.6152
447	.718	545.118	-.01	10.27	.9197	.1381	.0206	.0045	-.0003	6.657	.0043	161.2306
448	.717	544.359	-.00	11.39	.9415	.1681	.0455	.0055	-.0008	5.602	.0043	169.2306
449	.720	546.787	-.00	12.47	.9502	.1971	.0677	.0060	-.0013	4.822	.0042	162.4614
450	.718	545.442	-.00	13.58	.9426	.2202	.0938	.0016	-.0003	4.281	.0042	124.6153
451	.721	548.151	-.01	15.81	.9691	.2739	.1153	.0009	.0000	3.538	.0040	110.4614
452	.720	547.380	-.01	17.92	1.0003	.3243	.1329	.0013	-.0001	3.085	.0037	102.7691
453	.720	547.465	-.00	19.99	1.0348	.3805	.1345	.0011	-.0001	2.719	.0031	112.6153
454	.722	548.626	-.00	22.09	1.0783	.4430	.1367	.0007	-.0001	2.434	.0025	122.7691
455	.719	546.482	-.00	8.17	.8669	.0860	-.0470	.0002	-.0000	10.079	.0045	77.6922

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 69

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
469	.663	488.289	-.00	.18	.2074	.0191	.0013	.0006	.0004	.0042	0.0000
470	.664	489.418	-.01	1.26	.3010	.0159	.0013	.0007	.0013	.0043	0.0000
471	.667	491.755	-.00	2.34	.3936	.0101	.0016	.0007	.0006	.0043	0.0000
472	.665	489.979	-.01	3.45	.4914	.0015	.0015	.0007	.0015	.0043	0.0000
473	.665	489.500	-.01	4.55	.5790	-.0093	.0015	.0008	.0011	.0043	0.0000
474	.665	490.076	-.00	5.68	.6610	-.0215	.0019	.0007	.0000	.0043	0.0000
475	.665	489.539	-.01	6.81	.7462	-.0348	.0019	.0009	.0009	.0043	0.0000
476	.665	489.661	-.00	7.96	.8259	-.0410	.0020	.0005	.0016	.0042	0.0000
477	.666	491.190	-.00	9.07	.8742	-.0356	-.0009	-.0001	.0014	.0042	0.0000
478	.665	490.117	-.00	10.16	.8951	-.0305	.0014	.0004	-.0001	.0042	0.0000
479	.667	492.265	-.00	11.29	.9230	-.0212	.0018	.0002	.0015	.0042	0.0000
480	.667	491.928	-.00	12.33	.9397	-.0161	.0029	.0002	.0012	.0041	0.0000
481	.667	492.180	-.00	13.47	.9604	-.0113	.0018	.0003	.0014	.0041	0.0000
482	.666	491.521	-.00	15.66	1.0016	-.0052	.0013	.0002	.0025	.0040	0.0000
483	.668	492.795	-.00	17.79	1.0418	-.0011	.0011	.0003	.0021	.0038	0.0000
484	.668	492.648	-.00	19.85	1.0897	.0030	.0009	.0001	.0035	.0032	0.0000
485	.667	492.395	-.00	21.87	1.1310	.0047	.0009	-.0001	.0034	.0026	0.0000
486	.667	491.757	-.01	8.00	.8246	-.0409	.0020	.0007	.0015	.0042	0.0000

TEST= 778 RUN= 69

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
469	.663	488.289	-.00	.18	.2073	.0197	-.0906	.0013	.0006	10.506	.0042	8.6150
470	.664	489.418	-.01	1.26	.3006	.0225	-.0868	.0013	.0007	13.332	.0043	7.9228
471	.667	491.755	-.00	2.34	.3929	.0261	-.0836	.0016	.0006	15.041	.0043	9.6150
472	.665	489.979	-.01	3.45	.4905	.0311	-.0810	.0015	.0006	15.772	.0043	9.5535
473	.665	489.500	-.01	4.55	.5779	.0367	-.0766	.0015	.0006	15.748	.0043	9.0612
474	.665	490.076	-.00	5.68	.6599	.0440	-.0683	.0020	.0005	15.003	.0043	10.7842
475	.665	489.539	-.01	6.81	.7451	.0539	-.0607	.0020	.0006	13.832	.0043	10.9996
476	.665	489.661	-.00	7.96	.8236	.0738	-.0467	.0020	.0002	11.161	.0042	42.0000
477	.666	491.190	-.00	9.07	.8689	.1027	-.0089	-.0009	.0000	8.460	.0042	112.6153
478	.665	490.117	-.00	10.16	.8864	.1278	.0211	.0014	.0002	6.934	.0041	136.8460
479	.667	492.265	-.00	11.29	.9093	.1599	.0505	.0018	-.0002	5.686	.0041	146.7691
480	.667	491.928	-.00	12.33	.9214	.1850	.0767	.0029	-.0004	4.981	.0041	122.7691
481	.667	492.180	-.00	13.47	.9366	.2127	.0991	.0018	-.0001	4.403	.0040	111.4614
482	.666	491.521	-.00	15.66	.9658	.2654	.1252	.0013	-.0002	3.639	.0038	105.2307
483	.668	492.795	-.00	17.79	.9923	.3173	.1395	.0011	-.0000	3.128	.0037	97.3845
484	.668	492.648	-.00	19.85	1.0239	.3729	.1403	.0009	-.0002	2.746	.0030	108.4614
485	.667	492.395	-.00	21.87	1.0479	.4256	.1387	.0007	-.0004	2.462	.0024	119.9999
486	.667	491.757	-.01	8.00	.8223	.0742	-.0463	.0021	.0004	11.078	.0042	41.5384

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 77d RUN= 70

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
500	.225	72.847	-.00	.05	.1937	.0204	.0016	.0012	-.0003	.0036	0.0000
501	.225	72.749	-.00	.94	.2733	.0190	.0020	.0014	-.0010	.0036	0.0000
502	.224	72.555	-.00	1.91	.3379	.0166	.0016	.0009	-.0008	.0034	0.0000
503	.224	72.555	-.00	2.89	.4167	.0106	.0015	.0012	-.0015	.0037	0.0000
504	.224	72.555	.00	3.90	.4965	.0021	.0015	.0012	-.0121	.0037	0.0000
505	.225	73.243	.00	4.87	.5710	-.0070	.0014	.0007	-.0122	.0036	0.0000
506	.225	73.245	.00	5.93	.6337	-.0174	.0013	.0009	-.0126	.0036	0.0000
507	.224	72.560	.00	6.93	.7032	-.0284	.0021	.0011	-.0139	.0037	0.0000
508	.224	72.466	.00	7.99	.7679	-.0403	.0020	.0006	-.0144	.0037	0.0000
509	.224	72.380	.00	8.99	.8314	-.0444	.0001	-.0006	-.0134	.0034	0.0000
510	.224	72.400	.00	10.04	.8940	-.0420	-.0006	-.0010	-.0135	.0035	0.0000
511	.224	72.424	.00	11.09	.9243	-.0354	-.0006	-.0009	-.0145	.0035	0.0000
512	.224	72.449	.00	12.16	.9545	-.0285	-.0006	-.0008	-.0155	.0032	0.0000
513	.225	73.189	.00	14.34	1.0164	-.0171	.0001	.0001	-.0177	.0030	0.0000
514	.224	72.546	.00	16.36	1.0666	-.0105	.0001	.0002	-.0189	.0028	0.0000
515	.224	72.495	.00	18.38	1.1237	-.0065	.0001	.0006	-.0100	.0024	0.0000
516	.224	72.440	.00	20.38	1.1453	-.0024	.0000	.0006	-.0104	.0022	0.0000
517	.223	71.582	.00	7.89	.7597	-.0383	.0002	.0006	-.0130	.0035	0.0000
518	.225	72.950	.00	-.22	.1786	.0266	.0017	.0010	-.0100	.0036	0.0000

TEST= 77d RUN= 70

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
500	.225	72.847	-.00	.05	.1936	.0206	-.0757	.0016	.0012	9.398	.0036	1.2461
501	.225	72.749	-.00	.94	.2729	.0235	-.0734	.0020	.0014	11.594	.0036	1.5846
502	.224	72.555	-.00	1.91	.3371	.0278	-.0787	.0016	.0009	12.108	.0034	1.2000
503	.224	72.555	-.00	2.89	.4157	.0316	-.0650	.0016	.0011	13.141	.0036	.8769
504	.224	72.555	.00	3.90	.4952	.0358	-.0513	.0015	.0011	13.830	.0036	1.0153
505	.225	73.243	.00	4.87	.5695	.0415	-.0541	.0014	.0006	13.730	.0036	.8461
506	.225	73.245	.00	5.93	.6321	.0481	-.0480	.0014	.0008	13.136	.0036	1.0000
507	.224	72.560	.00	6.93	.7015	.0567	-.0358	.0022	.0009	12.377	.0036	.9846
508	.224	72.466	.00	7.99	.7660	.0668	-.0287	.0021	.0003	11.466	.0036	1.7384
509	.224	72.380	.00	8.99	.8281	.0860	-.0161	.0001	-.0006	9.633	.0034	11.6922
510	.224	72.400	.00	10.04	.8876	.1145	.0023	-.0008	-.0009	7.749	.0034	26.3077
511	.224	72.424	.00	11.09	.9138	.1431	.0343	-.0008	-.0008	6.387	.0034	33.3846
512	.224	72.449	.00	12.16	.9391	.1732	.0662	-.0008	-.0007	5.422	.0032	38.3846
513	.225	73.189	.00	14.34	.9889	.2351	.1169	.0001	.0001	4.206	.0029	39.0769
514	.224	72.546	.00	16.36	1.0264	.2903	.1491	.0001	.0002	3.535	.0027	37.7692
515	.224	72.495	.00	18.38	1.0684	.3481	.1764	.0002	.0005	3.069	.0023	31.6923
516	.224	72.440	.00	20.38	1.0745	.3966	.1884	.0003	.0006	2.709	.0020	26.5384
517	.223	71.582	.00	7.89	.7577	.0664	-.0253	.0003	.0006	11.415	.0034	1.6615
518	.225	72.950	.00	-.22	.1787	.0259	-.0770	.0016	.0010	6.906	.0036	.7538

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 71

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
532	.454	267.293	-.00	.11	.1935	.0196	.0015	.0009	.0027	.0039	0.0000
533	.455	267.478	-.00	1.21	.2709	.0168	.0013	.0008	.0025	.0039	0.0000
534	.455	268.019	-.00	2.20	.3517	.0116	.0009	.0009	.0022	.0040	0.0000
535	.455	267.569	-.00	3.27	.4299	.0046	.0009	.0007	.0018	.0040	0.0000
536	.455	267.570	-.00	4.31	.5072	-.0047	.0006	.0008	.0016	.0040	0.0000
537	.455	267.664	-.00	5.36	.5933	-.0156	.0011	.0007	.0032	.0040	0.0000
538	.455	267.578	-.00	6.49	.6623	-.0288	.0013	.0009	.0026	.0039	0.0000
539	.455	267.587	-.00	7.52	.7261	-.0417	.0004	.0007	.0028	.0039	0.0000
540	.455	267.541	-.00	8.65	.7955	-.0486	.0004	-.0006	.0049	.0039	0.0000
541	.455	268.265	-.00	9.71	.8516	-.0447	-.0008	-.0003	.0049	.0039	0.0000
542	.455	268.378	-.00	10.82	.8891	-.0357	.0004	.0003	.0028	.0038	0.0000
543	.455	267.846	-.00	11.89	.9159	-.0289	.0014	.0004	.0036	.0038	0.0000
544	.456	268.595	-.00	13.02	.9546	-.0238	.0014	.0005	.0026	.0037	0.0000
545	.456	268.632	-.00	15.21	1.0112	-.0153	.0008	.0002	.0045	.0036	0.0000
546	.456	268.828	-.01	17.34	1.0506	-.0102	.0008	.0004	.0063	.0035	0.0000
547	.456	268.846	-.01	19.35	1.0991	-.0061	.0008	.0006	.0056	.0032	0.0000
548	.456	269.607	-.00	21.42	1.1444	-.0020	.0008	.0003	.0054	.0026	0.0000
549	.454	267.020	-.00	.16	.1894	.0193	.0014	.0009	.0028	.0039	0.0000

TEST= 778 RUN= 71

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
532	.454	267.293	-.00	.11	.1935	.0199	-.0808	.0015	.0009	9.707	.0039	4.4306
533	.455	267.478	-.00	1.21	.2705	.0225	-.0786	.0013	.0008	12.035	.0039	4.1229
534	.455	268.019	-.00	2.20	.3510	.0251	-.0727	.0009	.0008	13.976	.0040	4.0306
535	.455	267.569	-.00	3.27	.4289	.0291	-.0691	.0009	.0007	14.747	.0040	4.6306
536	.455	267.570	-.00	4.31	.5061	.0335	-.0653	.0007	.0007	15.120	.0040	4.2768
537	.455	267.664	-.00	5.36	.5922	.0399	-.0575	.0012	.0006	14.841	.0040	4.7075
538	.455	267.578	-.00	6.49	.6614	.0463	-.0516	.0014	.0007	14.299	.0039	4.2768
539	.455	267.587	-.00	7.52	.7253	.0537	-.0414	.0005	.0007	13.498	.0039	4.7383
540	.455	267.541	-.00	8.65	.7938	.0716	-.0306	.0003	-.0006	11.091	.0039	9.1381
541	.455	268.265	-.00	9.71	.8469	.0996	-.0055	-.0008	-.0002	8.504	.0038	63.0769
542	.455	268.378	-.00	10.82	.8799	.1319	.0295	.0004	.0002	6.672	.0037	87.6922
543	.455	267.846	-.00	11.89	.9022	.1604	.0617	.0014	.0001	5.626	.0037	91.9999
544	.456	268.595	-.00	13.02	.9354	.1919	.0890	.0014	.0002	4.873	.0036	98.9230
545	.456	268.632	-.00	15.21	.9798	.2505	.1271	.0009	-.0000	3.912	.0035	102.4614
546	.456	268.828	-.01	17.34	1.0059	.3033	.1510	.0009	.0001	3.316	.0033	98.0768
547	.456	268.846	-.01	19.35	1.0390	.3584	.1682	.0010	.0003	2.899	.0031	95.7691
548	.456	269.607	-.00	21.42	1.0661	.4160	.1695	.0008	.0000	2.563	.0025	98.0768
549	.454	267.020	-.00	.16	.1894	.0198	-.0830	.0014	.0009	9.563	.0039	4.6460

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 72

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
586	.881	698.677	-.00	-.09	-.0349	.0170	.0004	.0007	-.0014	.0054	0.0000
587	.882	700.019	-.00	1.24	.1385	.0167	.0002	.0008	-.0019	.0055	0.0000
588	.883	700.429	-.00	2.65	.3153	.0154	.0002	.0007	-.0013	.0054	0.0000
589	.882	700.153	-.00	3.98	.4540	.0149	-.0006	.0007	-.0019	.0054	0.0000
590	.883	700.711	-.00	5.16	.5576	.0160	-.0014	.0006	-.0024	.0053	0.0000
591	.881	700.126	-.00	6.38	.6492	.0188	-.0036	.0006	-.0025	.0052	0.0000
592	.886	704.442	.00	7.62	.7246	.0193	-.0032	.0005	-.0027	.0052	0.0000
593	.884	702.629	-.00	8.80	.8003	.0195	-.0013	.0006	-.0024	.0050	0.0000
594	.886	704.331	.00	9.95	.8604	.0204	.0024	.0002	-.0018	.0050	0.0000
595	.888	705.299	.00	11.14	.8993	.0209	.0005	.0000	-.0007	.0050	0.0000

TEST= 778 RUN= 72

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	COB	PB-1
586	.881	698.677	-.00	-.09	-.0349	.0170	.0120	.0004	.0007	-2.045	.0054	12.9697
587	.882	700.019	-.00	1.24	.1381	.0197	.0077	.0002	.0008	7.021	.0055	16.3972
588	.883	700.429	-.00	2.65	.3142	.0300	-.0059	.0002	.0007	10.481	.0054	18.9889
589	.882	700.153	-.00	3.98	.4519	.0464	-.0108	-.0006	.0007	9.748	.0053	20.9084
590	.883	700.711	-.00	5.16	.5539	.0660	-.0081	-.0013	.0007	8.387	.0052	40.3771
591	.881	700.126	-.00	6.38	.6431	.0908	.0000	-.0035	.0010	7.084	.0052	79.2450
592	.886	704.442	.00	7.62	.7157	.1152	.0039	-.0031	.0009	6.211	.0051	93.5735
593	.884	702.629	-.00	8.80	.7879	.1416	.0132	-.0012	.0007	5.563	.0050	152.3911
594	.886	704.331	.00	9.95	.8440	.1688	.0249	.0024	-.0002	5.001	.0049	157.6679
595	.888	705.299	.00	11.14	.8783	.1943	.0360	.0005	-.0001	4.521	.0049	200.3085

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 73

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
609	.823	648.065	-.01	-.16	-.0379	.0158	.0000	.0008	.0001	.0049	0.0000
610	.823	648.064	-.01	-.16	-.0377	.0156	.0002	.0008	-.0001	.0048	0.0000
611	.822	647.727	-.01	1.08	.0959	.0146	-.0002	.0007	-.0001	.0048	0.0000
612	.823	648.950	-.00	2.39	.2401	.0097	-.0003	.0006	-.0007	.0048	0.0000
613	.822	647.682	-.00	3.70	.3853	.0031	-.0003	.0006	-.0012	.0048	0.0000
614	.823	648.457	-.00	4.98	.5274	.0002	-.0008	.0005	-.0012	.0048	0.0000
615	.822	648.345	-.00	6.29	.6502	.0005	-.0021	.0005	-.0013	.0047	0.0000
616	.822	648.368	-.00	7.44	.7281	.0021	-.0004	.0006	-.0018	.0047	0.0000
617	.824	649.700	.00	8.60	.7624	.0063	-.0009	.0000	-.0010	.0046	0.0000
618	.824	650.185	.00	9.68	.7921	.0101	-.0014	-.0000	-.0001	.0045	0.0000
619	.825	651.305	.00	10.76	.7905	.0150	-.0003	-.0000	-.0005	.0045	0.0000
620	.825	650.974	.00	11.74	.7830	.0224	-.0007	.0005	-.0024	.0044	0.0000
621	.822	647.233	-.00	-.16	-.0381	.0158	-.0004	.0008	-.0007	.0048	0.0000

TEST= 778 RUN= 73

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
609	.823	648.065	-.01	-.16	-.0378	.0159	.0085	.0000	.0008	-2.379	.0049	5.7852
610	.823	648.064	-.01	-.16	-.0377	.0157	.0091	.0002	.0008	-2.393	.0048	11.5712
611	.822	647.727	-.01	1.08	.0956	.0164	.0153	-.0002	.0007	5.815	.0048	11.1599
612	.823	648.950	-.00	2.39	.2395	.0197	.0227	-.0003	.0006	12.149	.0048	12.3390
613	.822	647.682	-.00	3.70	.3843	.0280	.0284	-.0003	.0006	13.707	.0048	15.8213
614	.823	648.457	-.00	4.98	.5254	.0460	.0291	-.0008	.0006	11.420	.0048	19.2631
615	.822	648.345	-.00	6.29	.6462	.0717	.0282	-.0020	.0008	9.008	.0047	48.9454
616	.822	648.368	-.00	7.44	.7217	.0964	.0410	-.0003	.0007	7.483	.0046	125.5859
617	.824	649.700	.00	8.60	.7529	.1202	.0588	-.0009	.0002	6.263	.0046	162.3294
618	.824	650.185	.00	9.68	.7791	.1431	.0741	-.0014	.0002	5.444	.0045	192.4935
619	.825	651.305	.00	10.76	.7738	.1623	.0901	-.0003	.0000	4.769	.0044	198.6632
620	.825	650.974	.00	11.74	.7620	.1813	.0950	-.0006	.0006	4.203	.0043	199.4858
621	.822	647.233	-.00	-.16	-.0381	.0160	.0085	-.0004	.0008	-2.387	.0048	11.2696

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 74

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
635	.769	597.093	-.01	-.09	-.0353	.0153	-.0000	.0008	.0001	.0046	0.0000
636	.770	597.535	-.00	1.13	.0872	.0144	.0003	.0007	-.0004	.0047	0.0000
637	.765	593.584	-.00	2.35	.2085	.0104	-.0005	.0006	-.0013	.0046	0.0000
638	.772	600.164	-.00	3.60	.3279	.0036	-.0006	.0005	-.0015	.0046	0.0000
639	.770	598.169	-.00	4.86	.4528	-.0041	-.0009	.0005	-.0016	.0046	0.0000
640	.771	599.162	-.00	6.18	.5801	-.0097	-.0003	.0006	-.0024	.0046	0.0000
641	.770	598.402	-.00	7.34	.6707	-.0085	.0010	.0006	-.0013	.0045	0.0000
642	.771	598.909	.00	8.50	.7122	-.0028	-.0004	.0002	-.0008	.0045	0.0000
643	.770	598.808	.00	9.51	.7302	.0055	-.0012	.0002	-.0018	.0044	0.0000
644	.771	599.071	.00	10.61	.7342	.0137	-.0023	-.0000	-.0008	.0043	0.0000
645	.771	599.514	.00	11.70	.7621	.0206	-.0014	.0004	-.0022	.0043	0.0000
646	.771	599.747	.00	12.77	.7824	.0243	-.0014	.0004	-.0025	.0042	0.0000
647	.772	599.901	.00	13.87	.8044	.0266	-.0015	.0005	-.0027	.0041	0.0000
648	.774	601.759	.00	16.14	.8599	.0275	-.0013	.0001	-.0026	.0038	0.0000
649	.776	604.193	.00	18.35	.9524	.0272	-.0010	.0004	-.0027	.0032	0.0000
650	.774	601.652	.00	20.50	1.0199	.0256	-.0007	.0001	-.0025	.0025	0.0000
651	.769	596.658	-.00	-.08	-.0313	.0155	.0001	.0008	-.0012	.0046	0.0000

TEST= 778 RUN= 74

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
635	.769	597.093	-.01	-.09	-.0352	.0154	.0072	-.0000	.0008	-2.288	.0046	10.0357
636	.770	597.535	-.00	1.13	.0869	.0161	.0156	.0003	.0007	5.403	.0047	9.9809
637	.765	593.584	-.00	2.35	.2079	.0189	.0244	-.0004	.0007	10.986	.0046	9.4325
638	.772	600.164	-.00	3.60	.3270	.0242	.0338	-.0006	.0006	13.528	.0046	10.2277
639	.770	598.169	-.00	4.86	.4515	.0343	.0435	-.0008	.0006	13.172	.0046	12.3116
640	.771	599.162	-.00	6.18	.5778	.0527	.0505	-.0002	.0006	10.958	.0045	19.8795
641	.770	598.402	-.00	7.34	.6663	.0773	.0610	.0010	.0005	8.624	.0045	77.7368
642	.771	598.909	.00	8.50	.7048	.1025	.0779	-.0004	.0002	6.876	.0044	139.8445
643	.770	598.808	.00	9.51	.7193	.1261	.0887	-.0011	.0004	5.704	.0043	179.7429
644	.771	599.071	.00	10.61	.7192	.1487	.0932	-.0023	.0004	4.838	.0043	194.1388
645	.771	599.514	.00	11.70	.7421	.1747	.1000	-.0013	.0007	4.248	.0042	227.8663
646	.771	599.747	.00	12.77	.7577	.1966	.1036	-.0013	.0007	3.854	.0041	257.8920
647	.772	599.901	.00	13.87	.7745	.2187	.1085	-.0013	.0008	3.542	.0040	262.8277
648	.774	601.759	.00	16.14	.8184	.2655	.1200	-.0012	.0005	3.083	.0036	185.0899
649	.776	604.193	.00	18.35	.8954	.3256	.1195	-.0008	.0007	2.750	.0030	170.2828
650	.774	601.652	.00	20.50	.9463	.3812	.1238	-.0006	.0004	2.483	.0024	201.5424
651	.769	596.658	-.00	-.08	-.0312	.0155	.0075	.0001	.0008	-2.009	.0046	9.5422

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 75

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
673	.717	543.173	-.00	-.18	-.0260	.0151	.0000	.0007	-.0000	.0044	0.0000
674	.717	543.793	-.00	.97	.0873	.0141	-.0000	.0007	-.0000	.0044	0.0000
675	.717	543.880	-.00	2.18	.1965	.0104	-.0001	.0006	-.0014	.0044	0.0000
676	.717	543.275	-.00	3.39	.3058	.0044	-.0001	.0005	-.0002	.0045	0.0000
677	.718	544.547	-.00	4.57	.4163	-.0036	.0000	.0005	-.0017	.0044	0.0000
678	.716	543.090	-.00	5.78	.5284	-.0105	-.0003	.0003	-.0002	.0044	0.0000
679	.718	544.359	.00	6.98	.6156	-.0107	.0003	.0004	-.0019	.0043	0.0000
680	.717	543.723	-.00	8.10	.6588	-.0016	.0019	.0008	-.0028	.0043	0.0000
681	.718	544.621	.00	9.20	.6953	.0058	-.0013	-.0001	-.0004	.0042	0.0000
682	.719	545.415	.00	10.26	.7226	.0114	-.0023	-.0000	-.0014	.0042	0.0000
684	.718	545.149	-.00	11.38	.7521	.0171	-.0014	.0004	-.0015	.0042	0.0000
685	.719	545.389	.00	12.43	.7746	.0205	-.0015	.0003	-.0019	.0041	0.0000
686	.718	545.096	.00	13.59	.7954	.0231	-.0014	.0003	-.0024	.0040	0.0000
687	.719	546.216	.00	15.79	.8404	.0257	-.0021	.0001	-.0024	.0037	0.0000
688	.720	547.020	-.00	17.98	.9161	.0260	-.0010	.0004	-.0017	.0031	0.0000
689	.721	547.367	.00	20.07	.9838	.0255	-.0006	.0001	-.0028	.0025	0.0000
690	.721	547.944	.00	22.20	1.0582	.0237	-.0010	-.0001	-.0032	.0017	0.0000

TEST= 778 RUN= 75

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
673	.717	543.173	-.00	-.18	-.0260	.0152	.0089	.0000	.0007	-1.706	.0044	7.8964
674	.717	543.793	-.00	.97	.0871	.0155	.0157	-.0000	.0007	5.606	.0044	7.7867
675	.717	543.880	-.00	2.18	.1959	.0178	.0246	-.0001	.0006	10.979	.0044	7.6496
676	.717	543.275	-.00	3.39	.3050	.0224	.0350	-.0000	.0005	13.589	.0044	7.8278
677	.718	544.547	-.00	4.57	.4152	.0296	.0451	.0001	.0005	14.020	.0044	8.8697
678	.716	543.090	-.00	5.78	.5268	.0428	.0555	-.0002	.0003	12.297	.0043	12.7777
679	.718	544.359	.00	6.98	.6123	.0642	.0648	.0003	.0003	9.531	.0043	53.0585
680	.717	543.723	-.00	8.10	.6525	.0912	.0764	.0019	.0005	7.153	.0043	141.2156
681	.718	544.621	.00	9.20	.6855	.1169	.0846	-.0013	.0001	5.863	.0042	172.2007
682	.719	545.415	.00	10.26	.7090	.1399	.0960	-.0023	.0004	5.066	.0042	176.8637
684	.718	545.149	-.00	11.38	.7340	.1652	.1068	-.0013	.0007	4.444	.0041	166.5810
685	.719	545.389	.00	12.43	.7520	.1867	.1136	-.0014	.0006	4.027	.0040	152.1851
686	.718	545.096	.00	13.59	.7678	.2093	.1203	-.0012	.0006	3.669	.0039	164.5244
687	.719	546.216	.00	15.79	.8017	.2534	.1304	-.0020	.0007	3.164	.0036	150.9511
688	.720	547.020	-.00	17.98	.8633	.3075	.1335	-.0008	.0007	2.807	.0030	142.7249
689	.721	547.367	.00	20.07	.9153	.3615	.1333	-.0005	.0003	2.532	.0023	159.1774
690	.721	547.944	.00	22.20	.9708	.4218	.1383	-.0010	.0003	2.302	.0016	178.0977

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 76

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
704	.665	489.530	-.00	-.15	-.0240	.0152	.0003	.0008	-.0003	.0043	0.0000
705	.665	490.095	-.00	1.04	.0830	.0143	.0001	.0007	.0000	.0042	0.0000
706	.663	488.008	-.00	2.15	.1814	.0110	-.0002	.0006	-.0010	.0043	0.0000
707	.665	489.232	-.00	3.33	.2838	.0053	.0002	.0006	-.0013	.0042	0.0000
708	.664	488.855	-.00	4.48	.3864	-.0023	-.0002	.0005	-.0007	.0042	0.0000
709	.665	490.128	-.00	5.66	.4878	-.0087	-.0001	.0004	-.0007	.0042	0.0000
710	.665	489.807	-.00	6.84	.5717	-.0067	-.0002	.0006	-.0003	.0042	0.0000
711	.666	490.497	-.00	7.94	.6332	-.0004	.0026	.0008	-.0023	.0041	0.0000
712	.666	490.875	-.00	9.04	.6709	.0064	.0011	.0005	-.0015	.0041	0.0000
713	.666	491.067	.00	10.12	.7077	.0103	-.0020	.0000	-.0009	.0041	0.0000
714	.666	491.267	.00	11.24	.7397	.0137	-.0016	.0003	-.0019	.0040	0.0000
715	.665	490.417	.00	12.32	.7688	.0169	-.0012	.0005	-.0029	.0040	0.0000
716	.666	491.195	-.00	13.45	.7921	.0203	-.0015	.0004	-.0016	.0039	0.0000
717	.667	491.500	.00	15.64	.8297	.0239	-.0016	.0005	-.0021	.0036	0.0000
718	.667	491.930	.00	17.76	.8912	.0248	-.0012	.0005	-.0032	.0031	0.0000
719	.667	492.222	.00	19.86	.9545	.0247	-.0013	.0005	-.0038	.0024	0.0000
720	.668	493.129	.00	21.97	1.0230	.0236	-.0013	.0001	-.0015	.0018	0.0000
721	.664	489.214	-.01	-.12	-.0241	.0151	.0003	.0008	.0012	.0042	0.0000

TEST= 778 RUN= 76

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
704	.665	489.530	-.00	-.15	-.0239	.0153	.0093	.0003	.0008	-1.567	.0043	6.7722
705	.665	490.095	-.00	1.04	.0827	.0158	.0165	.0001	.0007	5.253	.0042	6.9505
706	.663	488.008	-.00	2.15	.1809	.0177	.0255	-.0002	.0006	10.191	.0043	6.9230
707	.665	489.232	-.00	3.33	.2830	.0217	.0356	.0003	.0006	13.016	.0042	6.7037
708	.664	488.855	-.00	4.48	.3854	.0279	.0450	-.0002	.0005	13.798	.0042	7.1287
709	.665	490.128	-.00	5.66	.4863	.0395	.0550	-.0001	.0004	12.323	.0042	8.3077
710	.665	489.807	-.00	6.84	.5684	.0614	.0646	-.0001	.0006	9.261	.0042	55.8005
711	.666	490.497	-.00	7.94	.6271	.0871	.0723	.0027	.0004	7.197	.0041	119.2792
712	.666	490.875	-.00	9.04	.6615	.1118	.0826	.0011	.0003	5.920	.0040	171.5167
713	.666	491.067	.00	10.12	.6949	.1345	.0964	-.0019	.0004	5.169	.0040	173.1619
714	.666	491.267	.00	11.24	.7228	.1576	.1100	-.0016	.0006	4.587	.0039	153.4190
715	.665	490.417	.00	12.32	.7475	.1805	.1200	-.0011	.0007	4.140	.0039	157.1208
716	.666	491.195	-.00	13.45	.7656	.2040	.1269	-.0014	.0008	3.754	.0038	142.3136
717	.667	491.500	.00	15.64	.7926	.2466	.1396	-.0014	.0009	3.214	.0035	120.9254
718	.667	491.930	.00	17.76	.8412	.2955	.1422	-.0010	.0008	2.846	.0030	125.0385
719	.667	492.222	.00	19.86	.8893	.3475	.1414	-.0010	.0009	2.559	.0023	143.1362
720	.668	493.129	.00	21.97	.9399	.4046	.1422	-.0012	.0005	2.323	.0016	167.4036
721	.664	489.214	-.01	-.12	-.0241	.0151	.0085	.0003	.0008	-1.591	.0042	6.9916

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 77

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
735	.224	72.544	-.00	.02	-.0167	.0101	-.0003	.0007	.0002	.0036	0.0000
736	.225	73.234	-.00	1.01	.0768	.0110	-.0004	.0010	-.0009	.0036	0.0000
737	.225	73.235	-.00	1.98	.1391	.0091	-.0001	.0012	-.0021	.0036	0.0000
738	.224	72.549	-.00	2.99	.2192	.0071	-.0001	.0008	-.0032	.0037	0.0000
739	.225	73.141	.00	3.98	.3115	.0020	-.0002	.0004	-.0041	.0036	0.0000
740	.225	73.045	.00	4.99	.3904	-.0043	.0005	.0006	-.0057	.0036	0.0000
741	.225	73.048	-.00	6.02	.4668	-.0130	-.0002	.0009	.0036	.0036	0.0000
742	.225	72.958	.00	7.03	.5465	-.0162	-.0003	.0005	-.0073	.0036	0.0000
743	.225	72.977	.00	8.10	.6240	-.0116	-.0007	.0008	-.0080	.0036	0.0000
744	.225	72.901	.00	9.11	.7027	-.0050	-.0011	.0010	-.0085	.0034	0.0000
745	.225	72.823	.00	10.16	.7505	.0002	-.0011	.0005	-.0093	.0034	0.0000
746	.225	72.843	.00	11.17	.7813	.0058	-.0011	.0006	-.0098	.0035	0.0000
747	.225	72.960	.00	12.23	.7947	.0101	-.0015	.0007	-.0102	.0032	0.0000
748	.226	73.470	.00	7.98	.6196	-.0089	-.0010	.0008	-.0076	.0036	0.0000

TEST= 778 RUN= 77

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
735	.224	72.544	-.00	.02	-.0167	.0101	.0075	-.0003	.0007	-1.658	.0036	.4524
736	.225	73.234	-.00	1.01	.0766	.0124	.0144	-.0004	.0010	6.180	.0036	.0548
737	.225	73.235	-.00	1.98	.1387	.0139	.0247	-.0000	.0012	9.942	.0036	0.0000
738	.224	72.549	-.00	2.99	.2186	.0185	.0336	-.0001	.0008	11.814	.0036	.4661
739	.225	73.141	.00	3.98	.3107	.0236	.0345	-.0001	.0004	13.188	.0036	.0137
740	.225	73.045	.00	4.99	.3893	.0296	.0432	.0006	.0006	13.133	.0036	.0411
741	.225	73.048	-.00	6.02	.4656	.0361	.0520	-.0001	.0009	12.905	.0036	.0137
742	.225	72.958	.00	7.03	.5443	.0508	.0654	-.0002	.0005	10.714	.0036	2.7418
743	.225	72.977	.00	8.10	.6194	.0765	.0732	-.0006	.0008	8.100	.0036	14.2310
744	.225	72.901	.00	9.11	.6946	.1063	.0757	-.0009	.0012	6.535	.0034	33.7276
745	.225	72.823	.00	10.16	.7387	.1325	.0871	-.0010	.0007	5.573	.0034	45.6550
746	.225	72.843	.00	11.17	.7654	.1570	.0946	-.0010	.0008	4.874	.0034	44.1468
747	.225	72.960	.00	12.23	.7745	.1783	.1148	-.0013	.0010	4.344	.0032	41.8161
748	.226	73.470	.00	7.98	.6148	.0772	.0729	-.0009	.0009	7.966	.0036	14.9713

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 78

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
749	.456	268.647	-.00	-.22	-.0221	.0163	-.0001	.0007	.0001	.0039	0.0000
750	.455	268.013	-.00	.81	.0676	.0158	.0007	.0008	-.0017	.0039	0.0000
751	.456	268.739	-.00	1.85	.1561	.0131	-.0004	.0007	-.0020	.0039	0.0000
752	.455	267.475	.00	2.94	.2379	.0087	-.0003	.0006	-.0032	.0038	0.0000
753	.455	268.112	.00	4.00	.3223	.0020	-.0006	.0006	-.0040	.0038	0.0000
754	.456	268.750	.00	5.09	.4064	-.0070	-.0007	.0006	-.0050	.0038	0.0000
756	.457	269.586	.00	6.18	.4854	-.0142	-.0007	.0003	-.0034	.0038	0.0000
757	.457	270.191	.00	7.30	.5685	-.0132	-.0004	.0010	-.0049	.0038	0.0000
758	.454	266.575	-.00	8.37	.6408	-.0046	.0002	.0008	-.0036	.0038	0.0000
759	.455	267.929	.00	9.42	.6844	.0016	.0002	.0006	-.0042	.0037	0.0000
760	.456	268.659	.00	10.59	.7324	.0071	-.0006	.0008	-.0045	.0036	0.0000
761	.455	268.187	.00	11.66	.7414	.0106	-.0011	.0005	-.0044	.0035	0.0000
762	.456	268.815	.00	12.76	.7722	.0137	-.0018	.0004	-.0046	.0035	0.0000
763	.457	269.512	.00	14.89	.8105	.0176	-.0014	.0005	-.0033	.0034	0.0000
764	.457	270.312	.00	16.94	.8576	.0210	-.0007	.0007	-.0046	.0031	0.0000
765	.456	268.923	-.00	7.30	.5708	-.0135	-.0005	.0012	-.0022	.0038	0.0000

TEST= 778 RUN= 78

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
749	.456	268.647	-.00	-.22	-.0220	.0164	.0070	-.0001	.0007	-1.343	.0039	4.9489
750	.455	268.013	-.00	.81	.0676	.0167	.0159	.0007	.0008	4.043	.0039	4.3458
751	.456	268.739	-.00	1.85	.1556	.0181	.0234	-.0003	.0008	8.595	.0039	4.4143
752	.455	267.475	.00	2.94	.2372	.0209	.0333	-.0003	.0007	11.366	.0038	4.6199
753	.455	268.112	.00	4.00	.3214	.0245	.0410	-.0006	.0006	13.138	.0038	4.5925
754	.456	268.750	.00	5.09	.4054	.0291	.0474	-.0006	.0007	13.911	.0038	4.7296
756	.457	269.586	.00	6.18	.4841	.0382	.0570	-.0007	.0004	12.678	.0037	5.1820
757	.457	270.191	.00	7.30	.5656	.0591	.0691	-.0002	.0010	9.563	.0037	18.1663
758	.454	266.575	-.00	8.37	.6346	.0887	.0755	.0003	.0008	7.151	.0037	66.3574
759	.455	267.929	.00	9.42	.6749	.1136	.0824	.0002	.0006	5.939	.0036	106.1173
760	.456	268.659	.00	10.59	.7186	.1416	.0969	-.0004	.0009	5.074	.0036	119.5534
761	.455	268.187	.00	11.66	.7240	.1602	.1118	-.0010	.0007	4.518	.0035	123.9407
762	.456	268.815	.00	12.76	.7501	.1840	.1283	-.0017	.0008	4.078	.0034	112.1498
763	.457	269.512	.00	14.89	.7787	.2253	.1489	-.0012	.0009	3.457	.0033	80.6163
764	.457	270.312	.00	16.94	.8143	.2699	.1564	-.0005	.0009	3.017	.0029	74.3096
765	.456	268.923	-.00	7.30	.5679	.0592	.0695	-.0003	.0012	9.594	.0037	20.0172

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 77d RUN= 79

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
801	.881	696.667	-.00	-.09	-.0053	.0154	.0004	.0007	-.0006	.0054	0.0000
802	.882	697.882	-.01	.98	.0818	.0145	.0006	.0007	-.0004	.0055	0.0000
803	.883	698.531	-.01	.99	.0837	.0145	.0011	.0007	-.0009	.0055	0.0000
804	.881	696.924	-.01	2.10	.1792	.0111	.0009	.0007	-.0004	.0054	0.0000
805	.883	698.661	-.01	3.27	.2689	.0060	.0009	.0007	-.0003	.0055	0.0000
806	.881	697.037	-.01	4.38	.3587	-.0001	.0015	.0007	-.0005	.0054	0.0000
807	.883	698.444	-.01	5.54	.4668	-.0030	.0018	.0010	-.0014	.0054	0.0000
808	.882	698.039	-.01	6.72	.5502	-.0004	.0021	.0007	.0002	.0054	0.0000
809	.884	699.576	-.00	7.91	.6306	.0022	.0014	.0005	.0004	.0053	0.0000
810	.880	695.966	-.00	9.04	.6690	.0009	.0011	.0004	-.0002	.0052	0.0000
811	.885	700.673	-.01	10.20	.7464	.0067	.0037	.0007	-.0008	.0052	0.0000
812	.885	700.876	-.00	11.33	.7449	.0136	.0003	.0002	-.0003	.0051	0.0000
813	.885	700.688	-.00	12.43	.7718	.0154	.0001	.0002	-.0008	.0050	0.0000
814	.884	699.700	-.00	13.59	.8020	.0165	.0005	.0003	.0005	.0049	0.0000
815	.884	699.321	-.01	-.06	-.0055	.0150	.0002	.0007	-.0004	.0054	0.0000

TEST= 77d RUN= 79

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
801	.881	696.667	-.00	-.09	-.0053	.0154	.0056	.0004	.0007	-.345	.0054	9.7722
802	.882	697.882	-.01	.98	.0816	.0159	.0044	.0006	.0007	5.143	.0055	-.0659
803	.883	698.531	-.01	.99	.0834	.0159	.0057	.0011	.0007	5.248	.0055	9.9259
804	.881	696.924	-.01	2.10	.1787	.0176	.0066	.0009	.0007	10.138	.0054	10.3651
805	.883	698.661	-.01	3.27	.2681	.0213	.0107	.0010	.0007	12.560	.0055	13.4399
806	.881	697.037	-.01	4.38	.3577	.0273	.0110	.0015	.0006	13.113	.0054	17.2940
807	.883	698.444	-.01	5.54	.4649	.0421	.0101	.0019	.0009	11.055	.0054	21.1811
808	.882	698.039	-.01	6.72	.5464	.0639	.0263	.0022	.0005	8.546	.0054	35.5758
809	.884	699.576	-.00	7.91	.6243	.0889	.0527	.0014	.0003	7.019	.0053	30.2505
810	.880	695.966	-.00	9.04	.6606	.1060	.0859	.0012	.0002	6.235	.0051	44.1408
811	.885	700.673	-.01	10.20	.7334	.1387	.1112	.0038	.0001	5.287	.0051	49.8506
812	.885	700.876	-.00	11.33	.7277	.1596	.1621	.0004	.0002	4.561	.0050	57.8662
813	.885	700.688	-.00	12.43	.7504	.1812	.1892	.0001	.0002	4.142	.0049	51.6074
814	.884	699.700	-.00	13.59	.7757	.2045	.2097	.0006	.0001	3.794	.0047	54.5721
815	.884	699.321	-.01	-.06	-.0055	.0150	.0066	.0002	.0007	-.366	.0054	9.6844

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 80

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
830	.821	643.981	-.01	-.21	-.0099	.0154	-.0001	.0008	.0009	.0048	0.0000
831	.820	642.917	-.01	.85	.0759	.0147	.0006	.0009	.0006	.0049	0.0000
832	.821	644.551	-.01	1.98	.1646	.0114	.0009	.0008	.0006	.0049	0.0000
833	.823	645.630	-.01	3.10	.2454	.0066	.0008	.0008	.0007	.0049	0.0000
834	.822	645.176	-.01	4.20	.3290	.0005	.0014	.0009	.0015	.0048	0.0000
835	.822	645.558	-.01	5.36	.4232	-.0029	.0014	.0005	.0019	.0048	0.0000
836	.823	645.915	-.01	6.49	.5035	-.0018	.0015	.0006	.0015	.0048	0.0000
837	.822	645.304	-.01	7.67	.5823	-.0023	.0018	.0007	.0007	.0047	0.0000
838	.823	646.198	-.01	8.84	.6462	.0008	.0044	.0012	-.0013	.0047	0.0000
839	.823	647.505	-.01	9.96	.6898	.0057	.0018	.0006	.0011	.0046	0.0000
840	.825	649.066	-.01	11.12	.7335	.0076	.0019	.0005	.0026	.0045	0.0000
841	.826	649.904	-.01	12.20	.7698	.0100	.0005	.0004	.0021	.0044	0.0000
842	.825	649.351	-.00	13.36	.7995	.0128	.0003	.0004	.0018	.0043	0.0000
843	.822	645.692	-.01	-.21	-.0114	.0153	.0004	.0009	.0006	.0048	0.0000

TEST= 778 RUN= 80

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
830	.821	643.981	-.01	-.21	-.0099	.0154	.0057	-.0001	.0008	-.640	.0048	9.3550
831	.820	642.917	-.01	.85	.0756	.0158	.0061	.0006	.0009	4.795	.0049	9.1134
832	.821	644.551	-.01	1.98	.1641	.0171	.0084	.0009	.0008	9.617	.0049	8.6083
833	.823	645.630	-.01	3.10	.2447	.0198	.0141	.0009	.0008	12.334	.0048	9.6404
834	.822	645.176	-.01	4.20	.3281	.0246	.0188	.0014	.0008	13.337	.0048	11.1337
835	.822	645.558	-.01	5.36	.4216	.0366	.0170	.0014	.0004	11.527	.0048	18.8093
836	.823	645.915	-.01	6.49	.5004	.0551	.0353	.0016	.0004	9.075	.0048	26.4074
837	.822	645.304	-.01	7.67	.5774	.0755	.0632	.0019	.0004	7.651	.0047	31.1289
838	.823	646.198	-.01	8.84	.6384	.1001	.1015	.0045	.0005	6.378	.0046	38.9801
839	.823	647.505	-.01	9.96	.6785	.1249	.1434	.0019	.0003	5.432	.0046	40.4075
840	.825	649.066	-.01	11.12	.7183	.1469	.1699	.0020	.0002	4.825	.0044	46.7761
841	.826	649.904	-.01	12.20	.7503	.1725	.1971	.0005	.0002	4.350	.0043	47.1055
842	.825	649.351	-.00	13.36	.7749	.1972	.2202	.0003	.0003	3.930	.0042	47.9839
843	.822	645.692	-.01	-.21	-.0113	.0153	.0051	.0004	.0009	-.738	.0048	9.5087

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 81

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
857	.768	595.007	-.00	-.10	-.0141	.0151	.0005	.0008	-.0006	.0046	0.0000
858	.769	595.526	-.01	.95	.0644	.0146	.0007	.0008	.0006	.0047	0.0000
859	.769	595.170	-.00	2.03	.1488	.0117	.0006	.0007	-.0003	.0047	0.0000
860	.769	595.702	-.01	3.18	.2326	.0070	.0005	.0007	-.0001	.0047	0.0000
861	.770	596.243	-.00	4.23	.3086	.0011	.0005	.0005	-.0000	.0046	0.0000
862	.769	596.039	-.00	5.34	.3914	-.0040	.0010	.0004	.0010	.0046	0.0000
863	.770	596.695	-.00	6.49	.4765	-.0028	.0012	.0005	.0008	.0046	0.0000
864	.770	596.837	-.01	7.63	.5494	.0001	.0022	.0010	.0004	.0046	0.0000
865	.770	596.739	-.01	8.79	.6183	.0005	.0051	.0014	-.0014	.0045	0.0000
866	.770	597.018	-.01	9.89	.6668	.0024	.0045	.0010	.0007	.0044	0.0000
867	.770	596.748	-.00	11.07	.7134	.0052	.0006	.0004	.0018	.0044	0.0000
868	.771	598.102	-.01	12.18	.7552	.0068	.0006	.0004	.0024	.0043	0.0000
869	.772	598.369	-.00	13.35	.7944	.0087	.0003	.0004	.0022	.0041	0.0000
870	.772	599.011	-.00	15.61	.8710	.0115	.0003	.0005	.0014	.0040	0.0000
871	.770	596.476	-.01	-.10	-.0145	.0151	-.0000	.0009	.0010	.0046	0.0000

TEST= 778 RUN= 81

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	L/D	STAB.PRESS.COEFF CDB	PB-1
857	.768	595.007	-.00	-.10	-.0140	.0152	.0048	.0005	.0008	-.926	.0046	8.8718
858	.769	595.526	-.01	.95	.0642	.0156	.0059	.0007	.0008	4.108	.0047	8.9597
859	.769	595.170	-.00	2.03	.1483	.0169	.0072	.0007	.0007	8.768	.0047	8.7401
860	.769	595.702	-.01	3.18	.2318	.0198	.0146	.0005	.0007	11.681	.0047	8.3228
861	.770	596.243	-.00	4.23	.3077	.0239	.0210	.0005	.0005	12.898	.0046	8.7181
862	.769	596.039	-.00	5.34	.3901	.0324	.0204	.0010	.0003	12.023	.0046	12.9564
863	.770	596.695	-.00	6.49	.4738	.0511	.0312	.0012	.0004	9.266	.0046	40.7369
864	.770	596.837	-.01	7.63	.5446	.0731	.0664	.0023	.0007	7.451	.0046	36.2350
865	.770	596.739	-.01	8.79	.6110	.0950	.0950	.0052	.0006	6.432	.0044	40.4075
866	.770	597.018	-.01	9.89	.6565	.1169	.1288	.0046	.0002	5.617	.0043	46.7761
867	.770	596.748	-.00	11.07	.6991	.1421	.1735	.0007	.0003	4.921	.0043	40.5173
868	.771	598.102	-.01	12.18	.7368	.1660	.1980	.0007	.0003	4.439	.0042	43.5918
869	.772	598.369	-.00	13.35	.7709	.1918	.2210	.0004	.0003	4.019	.0040	46.5565
870	.772	599.011	-.00	15.61	.8358	.2455	.2595	.0004	.0004	3.405	.0038	54.9015
871	.770	596.476	-.01	-.10	-.0145	.0151	.0046	-.0000	.0009	-.959	.0046	9.0036

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 77a RUN= 82

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
885	.717	542.995	-.01	-.10	-.0110	.0151	.0007	.0009	.0005	.0044	0.0000
886	.717	542.698	-.01	.97	.0680	.0144	.0002	.0009	.0011	.0044	0.0000
887	.716	542.322	-.01	2.04	.1457	.0116	.0004	.0009	.0011	.0045	0.0000
888	.717	542.872	-.01	3.14	.2208	.0072	.0007	.0009	.0010	.0045	0.0000
889	.716	542.509	-.01	4.21	.2961	.0011	.0007	.0008	.0010	.0045	0.0000
890	.717	543.099	-.01	5.29	.3777	-.0047	.0011	.0006	.0022	.0044	0.0000
891	.717	542.791	-.01	6.42	.4620	-.0046	.0018	.0008	.0017	.0044	0.0000
892	.717	543.152	-.01	7.57	.5366	-.0026	.0043	.0018	-.0009	.0044	0.0000
893	.717	543.069	-.01	8.72	.6043	-.0010	.0048	.0015	-.0006	.0043	0.0000
894	.718	544.361	-.01	9.82	.6580	-.0004	.0063	.0015	-.0000	.0042	0.0000
895	.719	545.786	-.01	11.00	.7037	.0026	.0008	.0007	.0021	.0041	0.0000
896	.718	544.412	-.01	12.11	.7511	.0041	.0005	.0005	.0031	.0040	0.0000
897	.718	544.678	-.01	13.24	.7944	.0054	.0004	.0005	.0028	.0039	0.0000
898	.721	546.800	-.00	15.51	.8761	.0074	.0002	.0004	.0020	.0038	0.0000
899	.720	546.751	-.00	17.69	.9440	.0085	-.0002	.0005	.0017	.0036	0.0000
900	.721	547.688	-.00	19.75	.9956	.0089	.0001	.0005	.0007	.0035	0.0000
902	.717	543.014	-.01	-.07	-.0116	.0147	.0003	.0011	.0022	.0044	0.0000

TEST= 77a RUN= 82

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
885	.717	542.995	-.01	-.10	-.0110	.0152	.0047	.0007	.0009	-.725	.0044	7.7519
886	.717	542.698	-.01	.97	.0678	.0156	.0077	.0002	.0009	4.356	.0044	7.5982
887	.716	542.322	-.01	2.04	.1452	.0168	.0101	.0004	.0009	8.645	.0045	7.2468
888	.717	542.872	-.01	3.14	.2201	.0192	.0146	.0007	.0008	11.439	.0045	8.1691
889	.716	542.509	-.01	4.21	.2952	.0229	.0221	.0008	.0008	12.920	.0044	8.0813
890	.717	543.099	-.01	5.29	.3766	.0302	.0254	.0012	.0005	12.482	.0044	11.1996
891	.717	542.791	-.01	6.42	.4596	.0471	.0281	.0019	.0005	9.754	.0044	26.5172
892	.717	543.152	-.01	7.57	.5323	.0682	.0555	.0045	.0012	7.810	.0043	23.1682
893	.717	543.069	-.01	8.72	.5974	.0906	.0868	.0049	.0008	6.593	.0042	30.8546
894	.718	544.361	-.01	9.82	.6485	.1118	.1176	.0065	.0004	5.798	.0042	46.9957
895	.719	545.786	-.01	11.00	.6903	.1368	.1702	.0009	.0005	5.047	.0041	35.6860
896	.718	544.412	-.01	12.11	.7335	.1616	.1963	.0006	.0004	4.539	.0039	43.4820
897	.718	544.678	-.01	13.24	.7720	.1872	.2179	.0005	.0004	4.123	.0038	50.2898
898	.721	546.800	-.00	15.51	.8422	.2414	.2617	.0003	.0003	3.489	.0036	59.9524
899	.720	546.751	-.00	17.69	.8968	.2949	.2974	-.0001	.0006	3.041	.0034	65.3328
900	.721	547.688	-.00	19.75	.9340	.3448	.3277	.0003	.0005	2.709	.0033	57.9760
902	.717	543.014	-.01	-.07	-.0115	.0147	.0053	.0003	.0011	-.784	.0044	7.8397

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 83

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
916	.664	488.513	-.00	-.13	-.0172	.0151	.0002	.0008	-.0003	.0042	0.0000
917	.665	489.645	-.00	.90	.0592	.0148	.0004	.0008	-.0003	.0042	0.0000
918	.665	489.017	-.00	1.93	.1381	.0122	.0005	.0007	-.0002	.0043	0.0000
919	.664	488.542	-.00	3.05	.2121	.0077	.0005	.0007	-.0002	.0042	0.0000
920	.665	489.682	-.00	4.11	.2829	.0020	.0005	.0006	-.0001	.0042	0.0000
921	.664	488.497	-.00	5.16	.3578	-.0044	.0007	.0004	.0013	.0042	0.0000
922	.666	490.696	-.00	6.29	.4446	-.0056	.0014	.0005	-.0005	.0042	0.0000
923	.664	488.838	-.01	7.40	.5177	-.0038	.0037	.0014	-.0029	.0042	0.0000
924	.665	489.568	-.01	8.62	.5886	-.0037	.0065	.0020	-.0043	.0041	0.0000
925	.666	490.956	-.01	9.70	.6504	-.0027	.0075	.0018	-.0044	.0040	0.0000
926	.666	490.038	-.00	10.87	.6968	.0010	.0006	.0003	.0001	.0039	0.0000
927	.665	489.341	-.00	11.99	.7369	.0017	.0005	.0002	-.0003	.0038	0.0000
928	.666	490.120	-.00	13.10	.7865	.0025	.0002	.0003	.0010	.0038	0.0000
929	.668	492.119	-.00	15.45	.8708	.0040	.0004	.0002	.0012	.0036	0.0000
930	.666	490.642	-.00	17.57	.9515	.0053	.0000	.0003	.0007	.0033	0.0000
931	.667	491.907	-.00	19.69	1.0053	.0059	-.0002	.0003	.0000	.0032	0.0000
932	.667	491.639	-.00	21.71	1.0612	.0054	.0007	.0003	.0004	.0031	0.0000
933	.665	489.727	-.00	-.14	-.0173	.0153	-.0001	.0008	-.0001	.0042	0.0000

TEST= 778 RUN= 83

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
916	.664	488.513	-.00	-.13	-.0171	.0152	.0025	.0002	.0008	-1.128	.0042	6.8954
917	.665	489.645	-.00	.90	.0590	.0157	.0055	.0004	.0008	3.752	.0042	7.0492
918	.665	489.017	-.00	1.93	.1376	.0168	.0078	.0005	.0007	8.195	.0043	7.0711
919	.664	488.542	-.00	3.05	.2114	.0189	.0132	.0006	.0007	11.167	.0042	7.0272
920	.665	489.682	-.00	4.11	.2821	.0222	.0202	.0005	.0006	12.699	.0042	7.0272
921	.664	488.497	-.00	5.16	.3567	.0278	.0248	.0007	.0003	12.816	.0042	8.3448
922	.666	490.696	-.00	6.29	.4426	.0431	.0243	.0015	.0003	10.259	.0042	15.4821
923	.664	488.838	-.01	7.40	.5139	.0629	.0475	.0039	.0009	8.177	.0041	27.8897
924	.665	489.568	-.01	8.62	.5825	.0846	.0750	.0067	.0010	6.888	.0041	23.3329
925	.666	490.956	-.01	9.70	.6416	.1069	.1041	.0077	.0005	6.004	.0040	46.0075
926	.666	490.038	-.00	10.87	.6841	.1324	.1661	.0006	.0002	5.169	.0039	31.4037
927	.665	489.341	-.00	11.99	.7205	.1547	.1903	.0005	.0001	4.656	.0038	39.6389
928	.666	490.120	-.00	13.10	.7655	.1807	.2116	.0002	.0002	4.236	.0037	50.7290
929	.668	492.119	-.00	15.45	.8383	.2358	.2585	.0004	.0000	3.555	.0035	64.0702
930	.666	490.642	-.00	17.57	.9055	.2923	.3000	.0001	.0003	3.098	.0032	68.6820
931	.667	491.907	-.00	19.69	.9446	.3444	.3358	-.0001	.0004	2.743	.0030	61.5997
932	.667	491.639	-.00	21.71	.9839	.3976	.3536	.0008	.0000	2.475	.0028	54.1879
933	.665	489.727	-.00	-.14	-.0173	.0154	.0024	-.0001	.0008	-1.127	.0042	7.0272

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 84

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
947	.225	72.741	-.00	-.04	-.0162	.0128	.0000	.0007	.0001	.0034	0.0000
948	.225	72.841	-.00	.96	.0478	.0134	.0003	.0008	-.0002	.0034	0.0000
949	.225	72.841	-.00	1.90	.1115	.0112	.0003	.0010	-.0002	.0034	0.0000
950	.224	72.253	-.00	2.93	.1929	.0086	.0002	.0013	.0001	.0034	0.0000
951	.224	72.255	-.00	3.93	.2577	.0045	.0002	.0008	.0003	.0034	0.0000
952	.224	72.157	-.00	4.92	.3068	-.0021	.0009	.0008	-.0003	.0034	0.0000
953	.224	72.062	.00	5.97	.3736	-.0082	.0026	.0002	-.0017	.0034	0.0000
954	.224	72.071	-.00	7.00	.4518	-.0109	.0026	.0004	-.0013	.0034	0.0000
955	.224	72.082	-.00	8.05	.5119	-.0123	.0007	.0008	.0003	.0035	0.0000
956	.224	72.101	-.00	9.10	.6059	-.0097	.0032	.0015	-.0028	.0035	0.0000
957	.224	72.115	-.00	10.23	.6339	-.0091	.0028	.0016	-.0030	.0032	0.0000
958	.224	72.131	-.00	11.22	.6757	-.0071	-.0012	.0006	-.0004	.0032	0.0000
959	.224	72.148	-.00	12.39	.7366	-.0098	-.0001	.0007	-.0021	.0033	0.0000
960	.225	72.878	.00	14.50	.8502	-.0105	-.0001	.0003	-.0031	.0030	0.0000
961	.224	72.224	.00	16.61	.8996	-.0127	.0000	.0004	-.0040	.0026	0.0000
962	.223	71.481	.00	18.62	.9995	-.0149	.0000	-.0001	-.0050	.0019	0.0000
963	.223	71.533	.00	20.62	1.0841	-.0121	-.0001	.0001	-.0056	.0020	0.0000
964	.224	72.021	.00	20.67	1.0605	-.0129	-.0001	.0001	-.0055	.0017	0.0000
965	.224	71.958	-.00	-.03	-.0006	.0158	.0000	.0007	-.0002	.0037	0.0000

TEST= 778 RUN= 84

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-P
947	.225	72.741	-.00	-.04	-.0162	.0128	-.0070	.0000	.0007	-1.266	.0034	.6039
948	.225	72.841	-.00	.96	.0476	.0142	.0005	.0003	.0008	3.350	.0034	1.4164
949	.225	72.841	-.00	1.90	.1110	.0149	.0079	.0003	.0010	7.468	.0034	2.0862
950	.224	72.253	-.00	2.93	.1922	.0184	.0103	.0003	.0013	10.446	.0034	1.4603
951	.224	72.255	-.00	3.93	.2568	.0221	.0122	.0002	.0007	11.629	.0034	1.0102
952	.224	72.157	-.00	4.92	.3058	.0242	.0191	.0009	.0008	12.662	.0034	.7906
953	.224	72.062	.00	5.97	.3724	.0307	.0216	.0026	-.0001	12.140	.0034	1.5372
954	.224	72.071	-.00	7.00	.4497	.0443	.0181	.0026	.0001	10.152	.0034	3.8430
955	.224	72.082	-.00	8.05	.5085	.0595	.0247	.0008	.0007	8.543	.0034	5.4900
956	.224	72.101	-.00	9.10	.5998	.0862	.0677	.0034	.0010	6.954	.0034	9.2012
957	.224	72.115	-.00	10.23	.6255	.1036	.0903	.0031	.0011	6.036	.0032	11.9243
958	.224	72.131	-.00	11.22	.6642	.1245	.1299	-.0011	.0008	5.336	.0032	12.4733
959	.224	72.148	-.00	12.39	.7215	.1485	.1652	.0000	.0007	4.859	.0032	13.4399
960	.225	72.878	.00	14.50	.8257	.2027	.2152	-.0000	.0003	4.073	.0029	19.4350
961	.224	72.224	.00	16.61	.8657	.2450	.2489	.0001	.0003	3.534	.0025	26.8466
962	.223	71.481	.00	18.62	.9520	.3050	.2973	-.0000	-.0001	3.121	.0018	25.4743
963	.223	71.533	.00	20.62	1.0189	.3705	.3366	-.0001	.0001	2.750	.0018	12.7370
964	.224	72.021	.00	20.67	.9968	.3623	.3279	-.0001	.0001	2.751	.0016	30.0860
965	.224	71.958	-.00	-.03	-.0006	.0158	.0049	.0000	.0007	-.039	.0037	.8125

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 85

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
979	.455	267.071	-.00	.14	-.0089	.0147	.0006	.0007	-.0007	.0038	0.0000
980	.455	266.981	-.00	1.20	.0649	.0140	.0003	.0009	-.0003	.0038	0.0000
981	.455	267.076	-.00	2.16	.1348	.0119	.0002	.0008	-.0001	.0038	0.0000
982	.455	267.713	-.00	3.21	.1955	.0081	.0004	.0007	-.0002	.0038	0.0000
983	.455	267.085	-.00	4.21	.2701	.0029	.0006	.0008	-.0003	.0038	0.0000
984	.455	267.089	-.00	5.25	.3272	-.0043	.0007	.0003	-.0003	.0038	0.0000
985	.455	267.108	.00	6.37	.4021	-.0110	.0016	-.0003	.0018	.0038	0.0000
986	.455	267.060	-.00	7.43	.4840	-.0137	.0012	-.0002	.0024	.0037	0.0000
987	.455	267.758	-.00	8.49	.5512	-.0122	.0023	.0011	-.0016	.0037	0.0000
988	.456	267.923	-.00	9.61	.6115	-.0102	.0051	.0016	-.0047	.0036	0.0000
989	.455	267.380	-.00	10.75	.6778	-.0065	.0021	.0008	-.0006	.0036	0.0000
990	.455	266.907	-.00	11.79	.7193	-.0059	.0003	.0004	.0028	.0035	0.0000
991	.456	268.254	-.00	12.93	.7616	-.0064	.0001	.0006	.0024	.0034	0.0000
992	.456	268.345	-.00	15.14	.8626	-.0067	.0002	.0002	.0013	.0031	0.0000
993	.458	269.900	-.00	17.31	.9449	-.0067	.0001	.0001	.0030	.0029	0.0000
994	.456	268.736	-.00	19.35	1.0191	-.0063	.0002	.0003	.0022	.0026	0.0000
995	.457	269.048	-.00	21.40	1.0953	-.0056	.0000	.0003	.0013	.0023	0.0000
996	.456	267.885	-.00	.11	-.0131	.0144	.0006	.0007	-.0007	.0038	0.0000

TEST= 778 RUN= 85

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
979	.455	267.071	-.00	.14	-.0089	.0147	.0040	.0006	.0007	-.606	.0038	3.1952
980	.455	266.981	-.00	1.20	.0646	.0153	.0082	.0003	.0009	4.209	.0038	3.2940
981	.455	267.076	-.00	2.16	.1342	.0170	.0109	.0002	.0008	7.885	.0038	3.5246
982	.455	267.713	-.00	3.21	.1948	.0191	.0147	.0005	.0007	10.209	.0038	3.4697
983	.455	267.085	-.00	4.21	.2691	.0227	.0237	.0006	.0007	11.847	.0038	3.3050
984	.455	267.089	-.00	5.25	.3262	.0256	.0259	.0008	.0003	12.720	.0038	3.4697
985	.455	267.108	.00	6.37	.4008	.0337	.0260	.0016	-.0005	11.903	.0038	5.2375
986	.455	267.060	-.00	7.43	.4818	.0490	.0274	.0012	-.0003	9.833	.0037	9.0036
987	.455	267.758	-.00	8.49	.5469	.0693	.0482	.0025	.0007	7.890	.0036	15.6799
988	.456	267.923	-.00	9.61	.6046	.0921	.0799	.0052	.0007	6.568	.0036	19.1387
989	.455	267.380	-.00	10.75	.6672	.1200	.1322	.0022	.0004	5.558	.0035	36.6742
990	.455	266.907	-.00	11.79	.7054	.1412	.1685	.0004	.0004	4.994	.0034	24.0469
991	.456	268.254	-.00	12.93	.7437	.1641	.1918	.0002	.0005	4.533	.0033	30.8546
992	.456	268.345	-.00	15.14	.8344	.2189	.2405	.0003	.0002	3.812	.0030	48.0937
993	.458	269.900	-.00	17.31	.9041	.2748	.2869	.0001	.0000	3.290	.0027	51.7172
994	.456	268.736	-.00	19.35	.9636	.3317	.3264	.0003	.0002	2.905	.0025	52.4858
995	.457	269.048	-.00	21.40	1.0219	.3944	.3736	.0001	.0002	2.591	.0022	51.9368
996	.456	267.885	-.00	.11	-.0132	.0143	.0023	.0006	.0007	-.918	.0038	3.4038

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 86

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
37	.881	695.176	-.01	-.24	-.0064	.0170	-.0008	.0007	.0026	.0054	0.0000
38	.883	696.956	-.01	.95	.1198	.0159	-.0003	.0007	.0026	.0054	0.0000
39	.883	697.290	-.01	3.40	.3967	.0086	-.0003	.0007	.0027	.0054	0.0000
40	.882	696.642	-.01	4.63	.5168	.0081	.0002	.0012	.0019	.0053	0.0000
41	.884	698.284	-.01	5.78	.6066	.0093	.0006	.0009	.0019	.0053	0.0000
42	.885	699.065	-.01	6.98	.6805	.0115	-.0028	.0004	.0031	.0053	0.0000
43	.883	697.134	-.01	8.16	.7415	.0126	.0004	.0006	.0024	.0052	0.0000
44	.885	698.843	-.01	9.27	.7648	.0159	.0028	.0007	.0024	.0051	0.0000
45	.879	694.126	-.01	2.11	.2499	.0115	-.0004	.0007	.0031	.0053	0.0000
46	.883	696.881	-.01	-.32	-.0031	.0173	-.0006	.0008	.0034	.0054	0.0000

TEST= 778 RUN= 86

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
37	.881	695.176	-.01	-.24	-.0064	.0171	.0058	-.0008	.0007	-.373	.0054	10.3887
38	.883	696.956	-.01	.95	.1195	.0179	.0062	-.0002	.0007	6.669	.0054	12.2953
39	.883	697.290	-.01	3.40	.3955	.0322	-.0069	-.0002	.0007	12.295	.0054	23.3444
40	.882	696.642	-.01	4.63	.5144	.0497	-.0138	.0003	.0012	10.345	.0052	20.4111
41	.884	698.284	-.01	5.78	.6026	.0703	-.0078	.0007	.0009	8.570	.0053	39.3555
42	.885	699.065	-.01	6.98	.6741	.0941	.0129	-.0027	.0007	7.160	.0052	78.8332
43	.883	697.134	-.01	8.16	.7322	.1176	.0366	.0005	.0005	6.224	.0051	99.9165
44	.885	698.843	-.01	9.27	.7523	.1389	.0670	.0029	.0002	5.415	.0051	134.9331
45	.879	694.126	-.01	2.11	.2493	.0207	.0058	-.0003	.0007	12.046	.0053	17.9666
46	.883	696.881	-.01	-.32	-.0030	.0173	.0062	-.0006	.0008	-.175	.0054	10.4865

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 776 RUN= 87

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
60	.769	593.553	-.01	.03	-.0051	.0165	-.0006	.0008	.0015	.0046	0.0000
61	.768	593.127	-.01	2.33	.2006	.0115	-.0008	.0009	.0017	.0046	0.0000
62	.769	593.641	-.01	2.26	.1966	.0117	-.0007	.0008	.0017	.0046	0.0000
63	.770	594.770	-.01	3.52	.2974	.0053	-.0002	.0007	.0019	.0046	0.0000
64	.769	593.925	-.00	4.61	.3969	-.0023	-.0000	.0007	.0011	.0046	0.0000
65	.769	593.532	-.01	5.83	.5058	-.0078	.0004	.0014	-.0000	.0046	0.0000
66	.770	594.864	-.01	7.05	.5921	-.0090	.0012	.0012	.0010	.0046	0.0000
67	.770	594.711	-.00	8.12	.6508	-.0051	-.0002	.0005	.0013	.0046	0.0000
68	.772	596.530	-.01	9.29	.6918	.0010	-.0001	.0006	.0025	.0045	0.0000
69	.771	595.984	-.01	10.42	.7273	.0059	.0038	.0010	.0008	.0045	0.0000
70	.772	596.772	-.01	11.51	.7819	.0090	.0119	.0016	.0003	.0045	0.0000
71	.771	596.071	-.01	12.62	.8221	.0112	.0138	.0015	.0017	.0044	0.0000
72	.772	596.453	-.00	13.73	.8108	.0172	.0000	.0004	.0014	.0043	0.0000
73	.772	596.527	-.00	15.93	.8708	.0202	-.0005	.0005	.0024	.0042	0.0000
74	.773	597.742	-.00	18.14	.9258	.0210	-.0009	.0006	.0012	.0040	0.0000
75	.770	594.942	-.01	1.10	.0898	.0151	-.0005	.0008	.0021	.0046	0.0000
76	.768	592.441	-.01	-.06	-.0052	.0164	-.0006	.0008	.0015	.0046	0.0000
77	.769	593.714	-.01	2.25	.1931	.0114	-.0001	.0008	.0024	.0046	0.0000

TEST= 778 RUN= 87

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	L/D	STAB.PRESS.COEFF CDB	PB-1
60	.769	593.553	-.01	.03	-.0051	.0165	.0057	-.0006	.0008	-.312	.0046	12.2953
61	.768	593.127	-.01	2.33	.1999	.0196	.0147	-.0008	.0009	10.178	.0046	9.1665
62	.769	593.641	-.01	2.26	.1960	.0194	.0142	-.0007	.0009	10.105	.0046	8.7754
63	.770	594.770	-.01	3.52	.2965	.0236	.0192	-.0001	.0007	12.584	.0046	8.8976
64	.769	593.925	-.00	4.61	.3958	.0296	.0247	.0000	.0007	13.376	.0046	10.4865
65	.769	593.532	-.01	5.83	.5040	.0436	.0319	.0006	.0014	11.556	.0046	16.3530
66	.770	594.864	-.01	7.05	.5887	.0638	.0442	.0013	.0010	9.234	.0045	29.5166
67	.770	594.711	-.00	8.12	.6449	.0869	.0708	-.0001	.0005	7.423	.0045	68.3221
68	.772	596.530	-.01	9.29	.6825	.1126	.1023	-.0000	.0006	6.059	.0044	71.6221
69	.771	595.984	-.01	10.42	.7143	.1374	.1172	.0039	.0003	5.199	.0044	112.7498
70	.772	596.772	-.01	11.51	.7644	.1648	.1175	.0120	-.0008	4.638	.0044	92.5832
71	.771	596.071	-.01	12.62	.7998	.1905	.1287	.0138	-.0015	4.198	.0042	99.1832
72	.772	596.453	-.00	13.73	.7836	.2091	.1648	.0001	.0004	3.747	.0042	112.5665
73	.772	596.527	-.00	15.93	.8318	.2584	.1800	-.0004	.0006	3.219	.0040	73.8832
74	.773	597.742	-.00	18.14	.8733	.3082	.1827	-.0007	.0008	2.833	.0038	77.7332
75	.770	594.942	-.01	1.10	.0895	.0168	.0100	-.0005	.0008	5.326	.0046	9.2643
76	.768	592.441	-.01	-.06	-.0052	.0164	.0064	-.0006	.0008	-.314	.0046	8.7510
77	.769	593.714	-.01	2.25	.1925	.0189	.0131	-.0001	.0008	10.165	.0046	9.4354

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 88

					BODY AXIS COEFFICIENTS					BODY PRESS.COEFF	
POINT	MACH	Q	BETA	ALPHA	CNF	CAF	CLB	CNB	CSF	CAB	CAC
92	.717	541.400	-.01	.05	-.0121	.0161	-.0007	.0009	.0018	.0043	0.0000
93	.717	541.406	-.01	1.17	.0859	.0153	-.0005	.0008	.0024	.0044	0.0000
94	.717	542.190	-.01	2.29	.1753	.0115	-.0002	.0008	.0029	.0044	0.0000
95	.717	541.668	-.01	3.40	.2813	.0062	-.0008	.0008	.0028	.0044	0.0000
96	.717	541.687	-.01	4.59	.3643	-.0016	-.0008	.0007	.0022	.0044	0.0000
97	.716	541.293	-.01	5.70	.4647	-.0081	-.0005	.0009	.0014	.0043	0.0000
98	.716	541.111	-.01	6.87	.5480	-.0060	.0004	.0011	.0011	.0043	0.0000
99	.718	542.826	-.01	8.03	.6083	-.0002	.0028	.0012	.0019	.0043	0.0000
100	.717	541.849	-.01	9.18	.6751	.0018	.0098	.0020	.0002	.0043	0.0000
101	.718	542.169	-.01	10.32	.7221	.0038	.0116	.0021	.0005	.0042	0.0000
102	.719	543.513	-.01	11.39	.7720	.0065	.0135	.0020	.0008	.0042	0.0000
103	.718	542.670	-.01	12.43	.7739	.0128	.0008	.0007	.0033	.0042	0.0000
104	.718	542.997	-.00	13.62	.7930	.0147	-.0005	.0004	.0027	.0041	0.0000
105	.720	544.530	-.00	15.78	.8528	.0182	-.0007	.0005	.0022	.0041	0.0000
106	.720	544.629	-.01	17.92	.9211	.0193	-.0007	.0007	.0030	.0039	0.0000
107	.721	545.405	-.01	20.03	.9660	.0211	-.0005	.0013	.0001	.0032	0.0000
108	.721	545.895	-.01	22.09	1.0443	.0198	-.0007	.0011	-.0002	.0026	0.0000
109	.718	542.155	-.01	5.59	.4659	-.0081	-.0003	.0011	.0038	.0044	0.0000

TEST= 778 RUN= 88

					STABILITY AXIS COEFFICIENTS					STAB.PRESS.COEFF		
POINT	MACH	Q	BETA	ALPHA	CL	CD	CPM	CLS	CNS	L/D	CDB	PB-1
92	.717	541.400	-.01	.05	-.0121	.0161	.0056	-.0007	.0009	-.752	.0043	7.9321
93	.717	541.406	-.01	1.17	.0856	.0170	.0103	-.0005	.0008	5.027	.0044	8.0299
94	.717	542.190	-.01	2.29	.1747	.0185	.0142	-.0001	.0008	9.432	.0044	8.1887
95	.717	541.668	-.01	3.40	.2805	.0229	.0206	-.0008	.0008	12.267	.0044	8.3232
96	.717	541.687	-.01	4.59	.3633	.0276	.0260	-.0007	.0008	13.164	.0044	8.5554
97	.716	541.293	-.01	5.70	.4632	.0381	.0326	-.0004	.0010	12.171	.0043	10.7798
98	.716	541.111	-.01	6.87	.5448	.0596	.0471	.0006	.0010	9.136	.0043	38.0722
99	.718	542.826	-.01	8.03	.6024	.0848	.0735	.0029	.0008	7.105	.0043	52.0666
100	.717	541.849	-.01	9.18	.6661	.1094	.0899	.0100	.0004	6.089	.0042	57.9332
101	.718	542.169	-.01	10.32	.7098	.1331	.1034	.0118	.0000	5.333	.0042	122.8331
102	.719	543.513	-.01	11.39	.7556	.1587	.1186	.0136	-.0007	4.759	.0041	92.9498
103	.718	542.670	-.01	12.43	.7530	.1791	.1569	.0009	.0005	4.205	.0041	131.5109
104	.718	542.997	-.00	13.62	.7672	.2010	.1709	-.0004	.0005	3.818	.0040	77.6110
105	.720	544.530	-.00	15.78	.8157	.2495	.1852	-.0005	.0006	3.270	.0039	70.0332
106	.720	544.629	-.01	17.92	.8705	.3018	.1937	-.0005	.0009	2.884	.0037	69.2999
107	.721	545.405	-.01	20.03	.9003	.3507	.1841	-.0000	.0014	2.567	.0030	82.3165
108	.721	545.895	-.01	22.09	.9602	.4112	.1912	-.0002	.0013	2.335	.0024	92.5832
109	.718	542.155	-.01	5.59	.4645	.0373	.0328	-.0002	.0011	12.452	.0044	10.2909

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 89

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
123	.664	487.154	-.00	-.04	-.0108	.0165	-.0007	.0008	.0020	.0041	0.0000
124	.665	487.720	-.01	1.08	.0864	.0155	-.0001	.0007	.0023	.0042	0.0000
125	.662	484.982	-.00	2.22	.1697	.0122	-.0006	.0007	.0023	.0042	0.0000
126	.663	486.119	-.01	3.37	.2641	.0063	-.0004	.0007	.0030	.0042	0.0000
127	.664	486.779	-.00	4.41	.3540	-.0010	.0000	.0007	.0020	.0042	0.0000
128	.665	487.378	-.00	5.49	.4411	-.0083	.0002	.0006	.0027	.0042	0.0000
129	.665	487.620	-.01	6.73	.5398	-.0071	.0006	.0011	.0016	.0041	0.0000
130	.665	487.799	-.01	7.88	.5952	-.0031	.0038	.0015	.0009	.0041	0.0000
131	.665	487.498	-.01	8.96	.6549	-.0003	.0088	.0020	.0000	.0041	0.0000
132	.665	488.166	-.01	10.04	.7050	.0019	.0119	.0022	.0009	.0040	0.0000
133	.666	488.378	-.01	11.16	.7599	.0043	.0139	.0020	.0014	.0040	0.0000
134	.664	487.143	-.00	12.20	.7629	.0110	-.0002	.0004	.0026	.0039	0.0000
135	.666	489.546	-.00	13.46	.7909	.0130	-.0006	.0003	.0025	.0039	0.0000
136	.666	488.903	-.00	15.59	.8492	.0163	-.0007	.0005	.0019	.0038	0.0000
137	.667	490.487	-.00	17.83	.9084	.0177	-.0009	.0005	.0029	.0037	0.0000
138	.667	489.798	-.01	19.85	.9586	.0207	-.0007	.0011	.0013	.0031	0.0000
139	.669	492.501	-.01	21.89	1.0299	.0199	-.0003	.0009	.0019	.0025	0.0000
140	.665	488.530	-.01	6.75	.5318	-.0073	.0006	.0011	.0016	.0041	0.0000

TEST= 778 RUN= 89

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
123	.664	487.154	-.00	-.04	-.0108	.0165	.0056	-.0007	.0008	-.656	.0041	6.3188
124	.665	487.720	-.01	1.08	.0861	.0171	.0113	-.0001	.0007	5.021	.0042	6.8199
125	.662	484.982	-.00	2.22	.1691	.0187	.0138	-.0006	.0007	9.023	.0042	6.7710
126	.663	486.119	-.01	3.37	.2633	.0218	.0209	-.0004	.0007	12.059	.0042	6.5510
127	.664	486.779	-.00	4.41	.3531	.0263	.0266	.0001	.0007	13.433	.0042	6.7710
128	.665	487.378	-.00	5.49	.4399	.0339	.0338	.0003	.0006	12.963	.0041	7.1132
129	.665	487.620	-.01	6.73	.5370	.0562	.0464	.0007	.0010	9.552	.0041	25.3611
130	.665	487.799	-.01	7.88	.5900	.0785	.0668	.0039	.0010	7.512	.0041	39.9055
131	.665	487.498	-.01	8.96	.6470	.1017	.0860	.0090	.0006	6.360	.0040	48.3999
132	.665	488.166	-.01	10.04	.6939	.1249	.1015	.0121	.0001	5.557	.0039	61.2332
133	.666	488.378	-.01	11.16	.7447	.1514	.1159	.0140	-.0007	4.920	.0039	76.6332
134	.664	487.143	-.00	12.20	.7434	.1720	.1579	-.0001	.0005	4.322	.0038	78.9554
135	.666	489.546	-.00	13.46	.7662	.1967	.1705	-.0005	.0005	3.894	.0038	73.1499
136	.666	488.903	-.00	15.59	.8136	.2439	.1883	-.0006	.0007	3.336	.0037	71.4999
137	.667	490.487	-.00	17.83	.8594	.2949	.1999	-.0007	.0008	2.914	.0035	66.9165
138	.667	489.798	-.01	19.85	.8946	.3449	.1903	-.0003	.0013	2.594	.0029	80.6665
139	.669	492.501	-.01	21.89	.9482	.4026	.1944	.0000	.0009	2.355	.0023	91.2998
140	.665	488.530	-.01	6.75	.5290	.0552	.0455	.0007	.0010	9.587	.0041	24.1388

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 77δ RUN= 90

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
154	.821	642.514	-.01	-.09	-.0068	.0165	-.0008	.0008	.0016	.0048	0.0000
155	.822	643.009	-.01	1.04	.0991	.0154	-.0008	.0007	.0022	.0048	0.0000
156	.820	641.678	-.01	2.18	.2073	.0111	-.0004	.0007	.0024	.0048	0.0000
157	.822	643.668	-.01	3.40	.3251	.0042	-.0002	.0007	.0016	.0047	0.0000
158	.822	643.248	-.01	4.57	.4363	-.0026	-.0003	.0008	.0022	.0047	0.0000
159	.822	643.042	-.01	5.79	.5555	-.0062	.0008	.0010	.0017	.0047	0.0000
160	.823	644.562	-.01	7.02	.6510	-.0037	.0019	.0010	.0010	.0047	0.0000
161	.823	644.305	-.00	9.24	.7168	.0072	.0016	.0006	.0015	.0046	0.0000

TEST= 77δ RUN= 90

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
154	.821	642.514	-.01	-.09	-.0067	.0166	.0057	-.0008	.0008	-.408	.0048	9.5576
155	.822	643.009	-.01	1.04	.0988	.0172	.0085	-.0007	.0007	5.749	.0048	9.2887
156	.820	641.678	-.01	2.18	.2067	.0190	.0102	-.0004	.0007	10.869	.0048	9.5820
157	.822	643.668	-.01	3.40	.3243	.0235	.0150	-.0001	.0007	13.823	.0047	12.4909
158	.822	643.248	-.01	4.57	.4351	.0322	.0170	-.0003	.0008	13.514	.0047	16.1575
159	.822	643.042	-.01	5.79	.5533	.0499	.0206	.0009	.0009	11.084	.0047	23.3444
160	.823	644.562	-.01	7.02	.6466	.0758	.0370	.0020	.0007	8.525	.0046	54.5110
161	.823	644.305	-.00	9.24	.7063	.1222	.0980	.0017	.0003	5.782	.0046	100.6498

TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Continued

TEST= 778 RUN= 91

POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
175	.450	268.142	-.00	.00	-.0011	.0175	-.0004	.0008	.0029	.0038	0.0000
176	.454	266.149	-.00	1.09	.0640	.0162	-.0002	.0006	.0024	.0038	0.0000
177	.456	267.418	-.00	2.14	.1497	.0133	-.0005	.0005	.0020	.0038	0.0000
178	.456	268.055	-.00	3.17	.2395	.0086	-.0000	.0006	.0037	.0038	0.0000
179	.456	267.333	-.00	4.26	.3047	.0021	-.0002	.0006	.0034	.0038	0.0000
180	.456	267.971	-.00	5.36	.3858	-.0070	-.0001	.0005	.0028	.0038	0.0000
181	.456	267.983	-.00	6.39	.4714	-.0163	-.0004	.0006	.0024	.0037	0.0000
182	.456	268.051	-.00	7.56	.5572	-.0145	-.0001	.0008	.0015	.0037	0.0000
183	.456	267.490	-.00	8.64	.6067	-.0105	.0026	.0016	-.0015	.0037	0.0000
184	.450	261.573	-.00	10.77	.7004	-.0009	-.0001	.0003	.0016	.0035	0.0000
185	.456	267.280	.00	13.02	.7866	.0031	-.0005	.0005	.0004	.0034	0.0000
186	.456	268.092	-.00	17.33	.9165	.0096	-.0009	.0006	.0015	.0031	0.0000
187	.456	267.644	-.00	21.26	1.0185	.0162	-.0008	.0009	.0010	.0025	0.0000

TEST= 778 RUN= 91

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
175	.450	268.142	-.00	.00	-.0011	.0175	.0030	-.0004	.0008	-.065	.0038	3.4466
176	.454	266.149	-.00	1.09	.0637	.0174	.0071	-.0002	.0006	3.658	.0038	3.2388
177	.456	267.418	-.00	2.14	.1491	.0189	.0136	-.0005	.0005	7.901	.0038	3.2388
178	.456	268.055	-.00	3.17	.2387	.0218	.0220	-.0000	.0006	10.927	.0038	3.6666
179	.456	267.333	-.00	4.26	.3037	.0248	.0262	-.0001	.0007	12.266	.0038	3.4099
180	.456	267.971	-.00	5.36	.3848	.0291	.0306	-.0001	.0005	13.244	.0038	3.4099
181	.456	267.983	-.00	6.39	.4703	.0363	.0385	-.0003	.0006	12.959	.0037	3.7277
182	.456	268.051	-.00	7.56	.5543	.0589	.0477	.0001	.0008	9.409	.0037	13.8842
183	.456	267.490	-.00	8.64	.6014	.0807	.0653	.0028	.0012	7.450	.0036	30.4333
184	.450	261.573	-.00	10.77	.6883	.1300	.1273	-.0000	.0003	5.294	.0035	47.9110
185	.456	267.280	.00	13.02	.7657	.1803	.1645	-.0004	.0006	4.248	.0033	55.1221
186	.456	268.092	-.00	17.33	.8720	.2822	.2121	-.0007	.0008	3.091	.0030	56.7110
187	.456	267.644	-.00	21.26	.9433	.3843	.2122	-.0005	.0011	2.454	.0023	65.6332

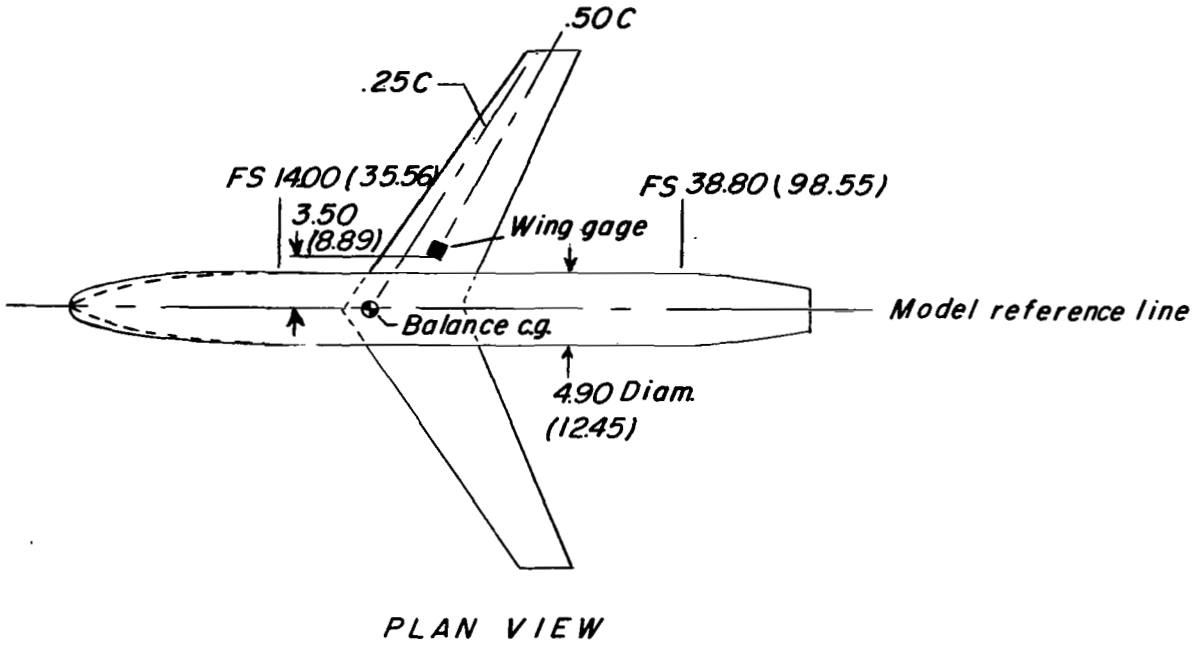
TABLE III.- BUFFET AND STATIC AERODYNAMIC RESULTS – Concluded

TEST= 778 RUN= 92

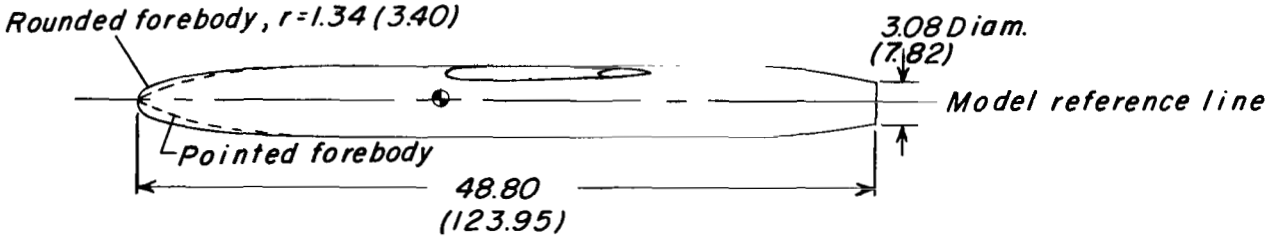
POINT	MACH	Q	BETA	ALPHA	BODY AXIS COEFFICIENTS CNF	CAF	CLB	CNB	CSF	BODY PRESS.COEFF CAB	CAC
188	.223	71.565	.00	.05	-.0163	.0183	-.0004	-.0000	.0004	.0035	0.0000
189	.224	72.254	.00	1.00	.0637	.0192	-.0000	.0002	-.0003	.0037	0.0000
190	.224	72.254	.00	1.97	.1275	.0166	.0003	.0004	-.0011	.0037	0.0000
191	.225	72.942	.00	2.97	.2054	.0116	.0002	-.0001	-.0016	.0036	0.0000
192	.225	72.845	.00	3.98	.2838	.0067	-.0005	.0002	-.0016	.0036	0.0000
193	.225	72.846	.00	4.98	.3468	-.0011	-.0002	-.0002	.0077	.0034	0.0000
194	.226	73.538	.00	6.10	.4681	-.0101	-.0003	.0002	.0066	.0034	0.0000
195	.226	73.542	.00	7.17	.4995	-.0163	-.0003	.0003	.0065	.0034	0.0000
196	.225	72.871	.00	8.20	.5683	-.0124	.0000	.0003	-.0040	.0034	0.0000
197	.225	72.912	.00	10.27	.7232	-.0041	-.0001	.0003	.0036	.0034	0.0000
198	.225	72.848	.00	12.50	.7849	-.0031	-.0002	.0005	.0023	.0032	0.0000
199	.224	72.201	.00	14.73	.8486	.0020	-.0012	-.0000	.0023	.0030	0.0000
200	.223	71.550	.00	16.67	.9143	.0051	-.0022	.0002	.0025	.0029	0.0000
201	.221	70.264	.00	20.75	1.0665	.0092	-.0014	.0006	.0000	.0022	0.0000

TEST= 778 RUN= 92

POINT	MACH	Q	BETA	ALPHA	STABILITY AXIS COEFFICIENTS CL	CD	CPM	CLS	CNS	STAB.PRESS.COEFF L/D	CDB	PB-1
188	.223	71.565	.00	.05	-.0163	.0183	-.0018	-.0004	-.0000	-.893	.0035	.6722
189	.224	72.254	.00	1.00	.0634	.0203	.0014	-.0000	.0002	3.128	.0037	1.0755
190	.224	72.254	.00	1.97	.1269	.0210	.0085	.0003	.0004	6.048	.0037	1.1733
191	.225	72.942	.00	2.97	.2045	.0222	.0172	.0002	-.0001	9.203	.0036	1.0022
192	.225	72.845	.00	3.98	.2827	.0264	.0259	-.0005	.0002	10.705	.0036	1.0633
193	.225	72.846	.00	4.98	.3456	.0290	.0272	-.0002	-.0001	11.905	.0034	1.3078
194	.226	73.538	.00	6.10	.4666	.0396	.0464	-.0003	.0002	11.775	.0034	1.2589
195	.226	73.542	.00	7.17	.4977	.0461	.0433	-.0003	.0003	10.787	.0034	2.7011
196	.225	72.871	.00	8.20	.5643	.0688	.0499	.0000	.0003	8.207	.0034	11.4153
197	.225	72.912	.00	10.27	.7123	.1249	.1106	-.0000	.0003	5.703	.0034	21.2055
198	.225	72.848	.00	12.50	.7669	.1669	.1493	-.0001	.0005	4.595	.0032	21.6944
199	.224	72.201	.00	14.73	.8201	.2178	.1757	-.0012	.0003	3.766	.0029	23.8333
200	.223	71.550	.00	16.67	.8744	.2671	.1917	-.0021	.0008	3.274	.0027	22.2444
201	.221	70.264	.00	20.75	.9941	.3864	.2364	-.0011	.0010	2.572	.0021	18.7000



PLAN VIEW



SIDE VIEW

Figure 1.- Sketches of typical buffet model. All linear dimensions are in inches (cm).

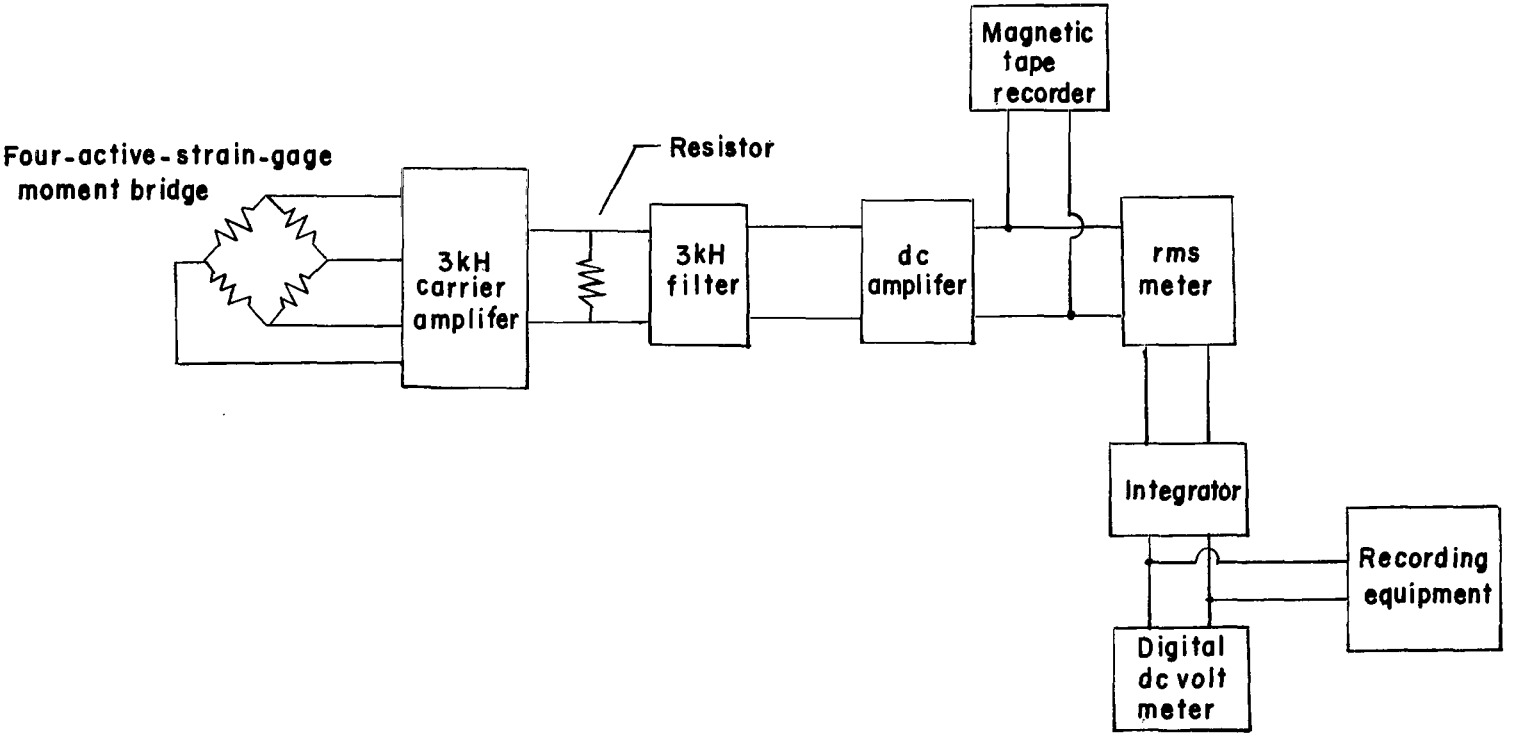
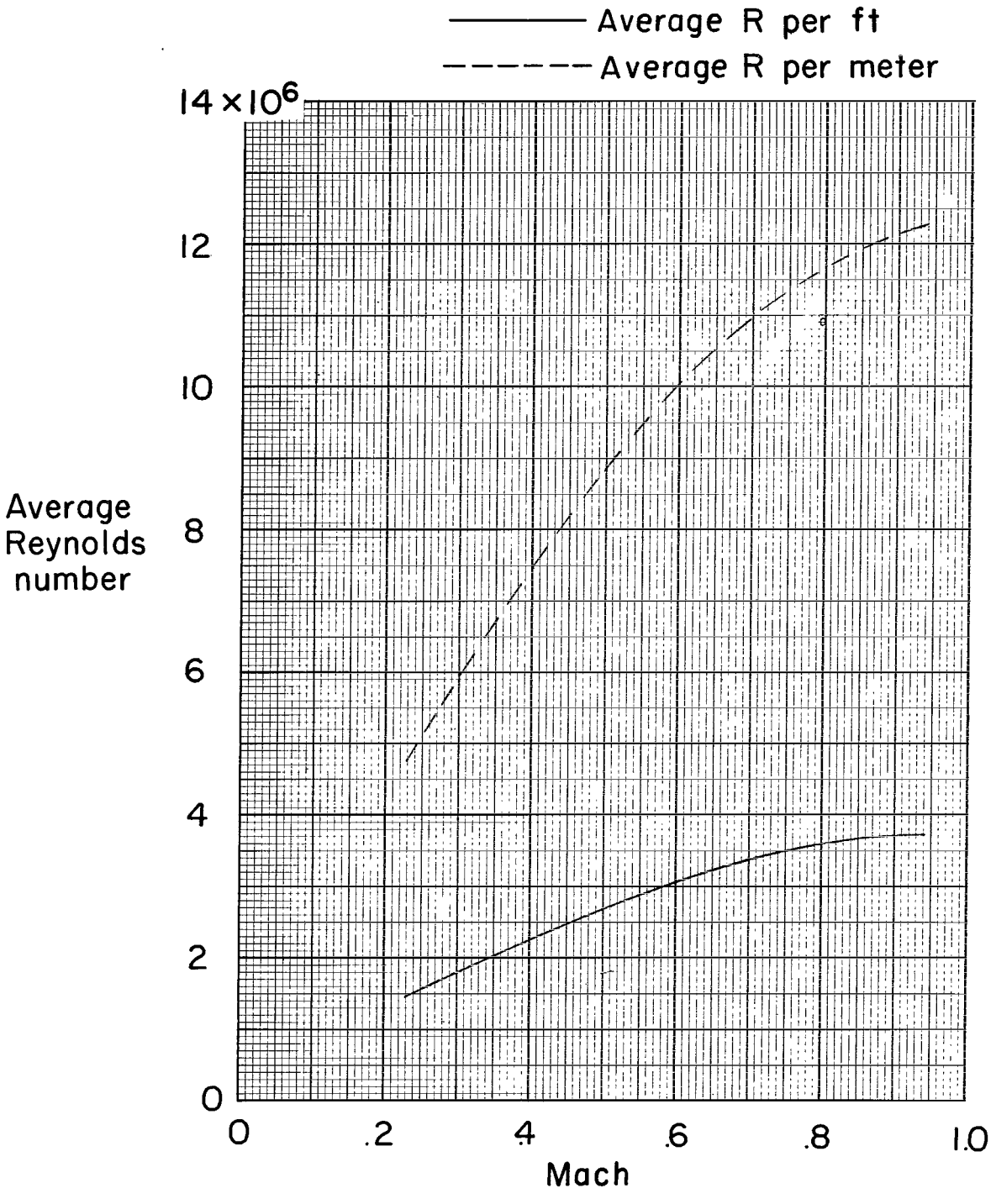
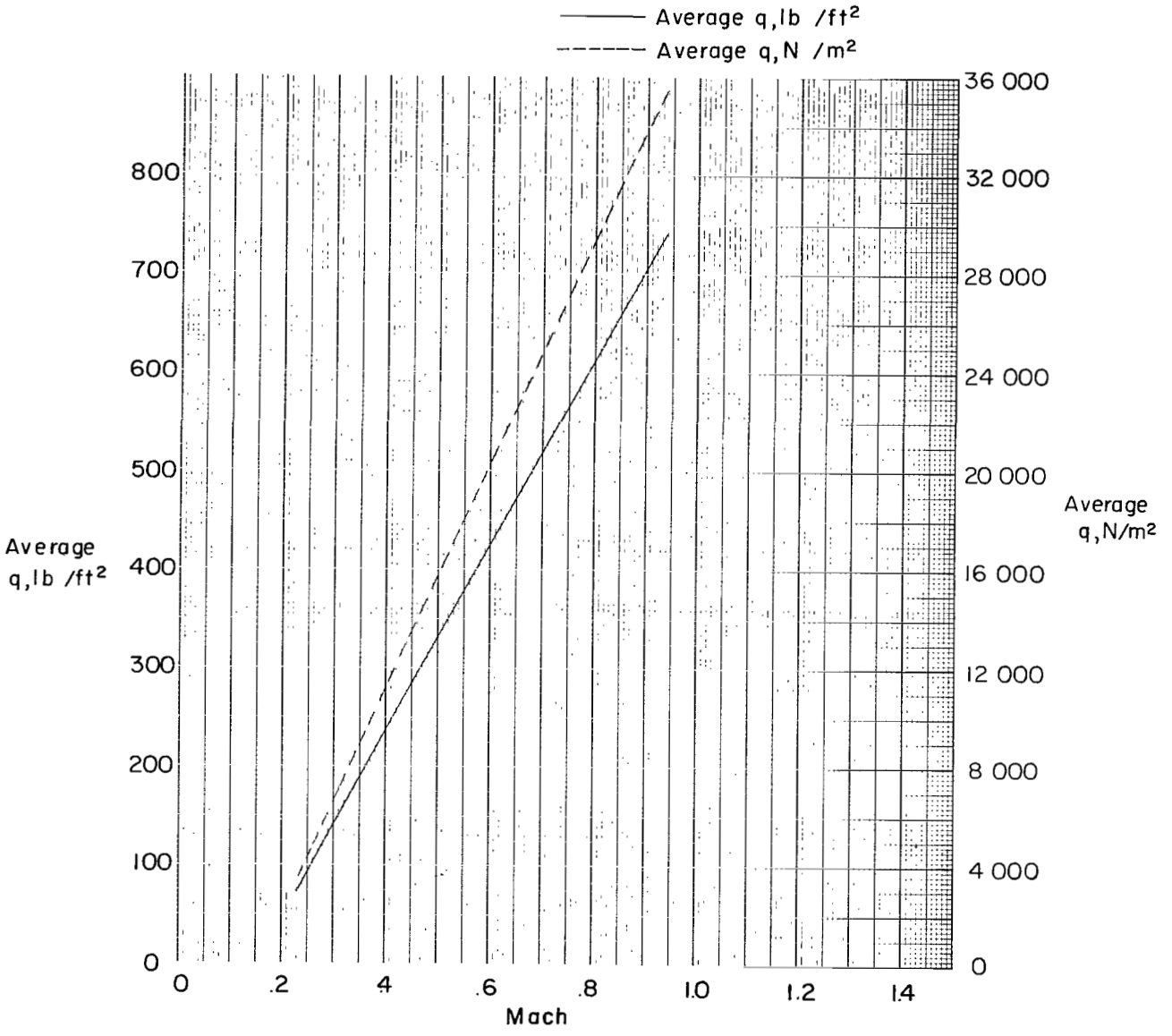


Figure 2.- Schematic drawing of wing-bending-gage electronic equipment and wiring arrangement.



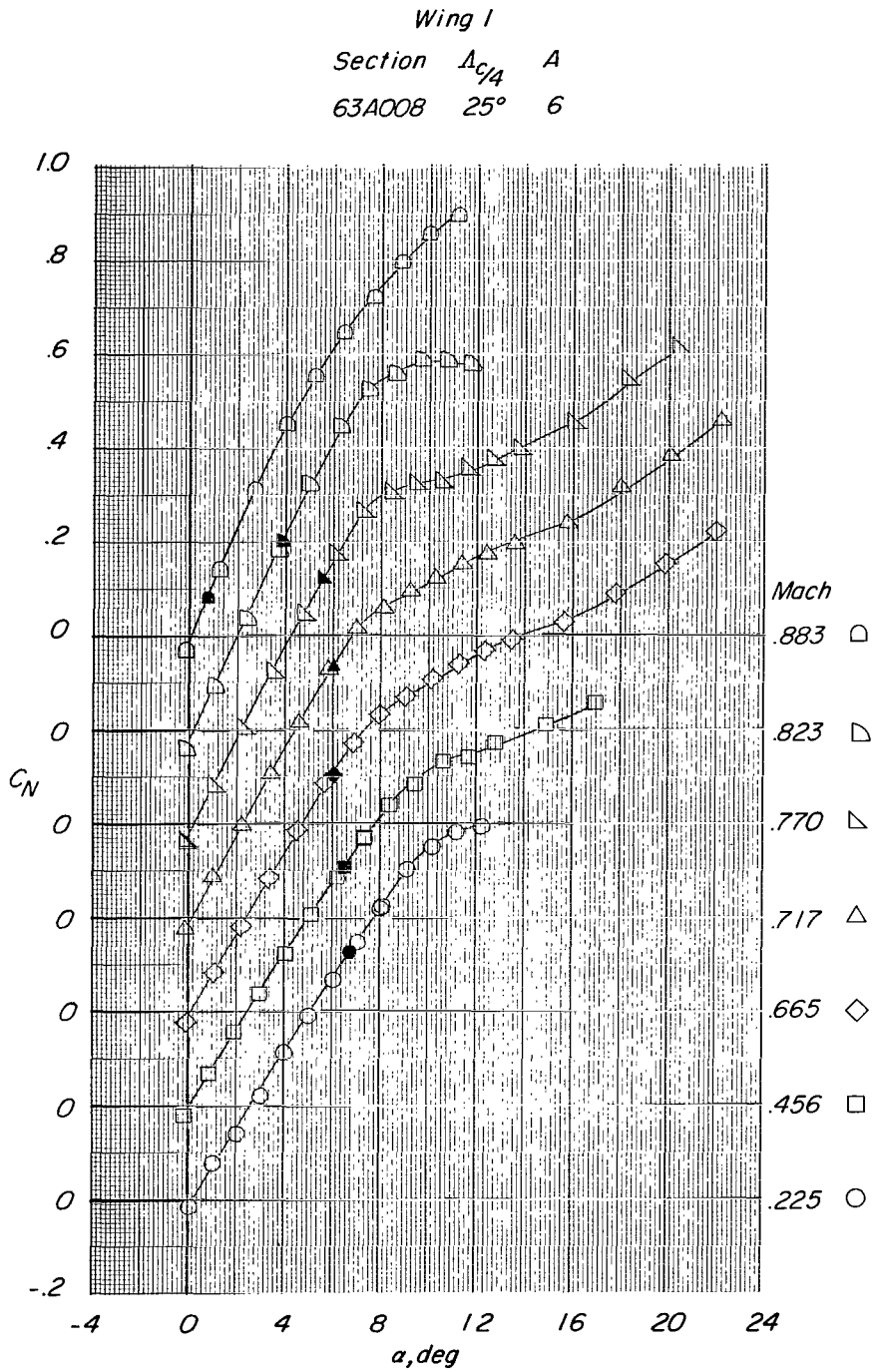
(a) Variation of Reynolds number with Mach number.

Figure 3.- Average test conditions.



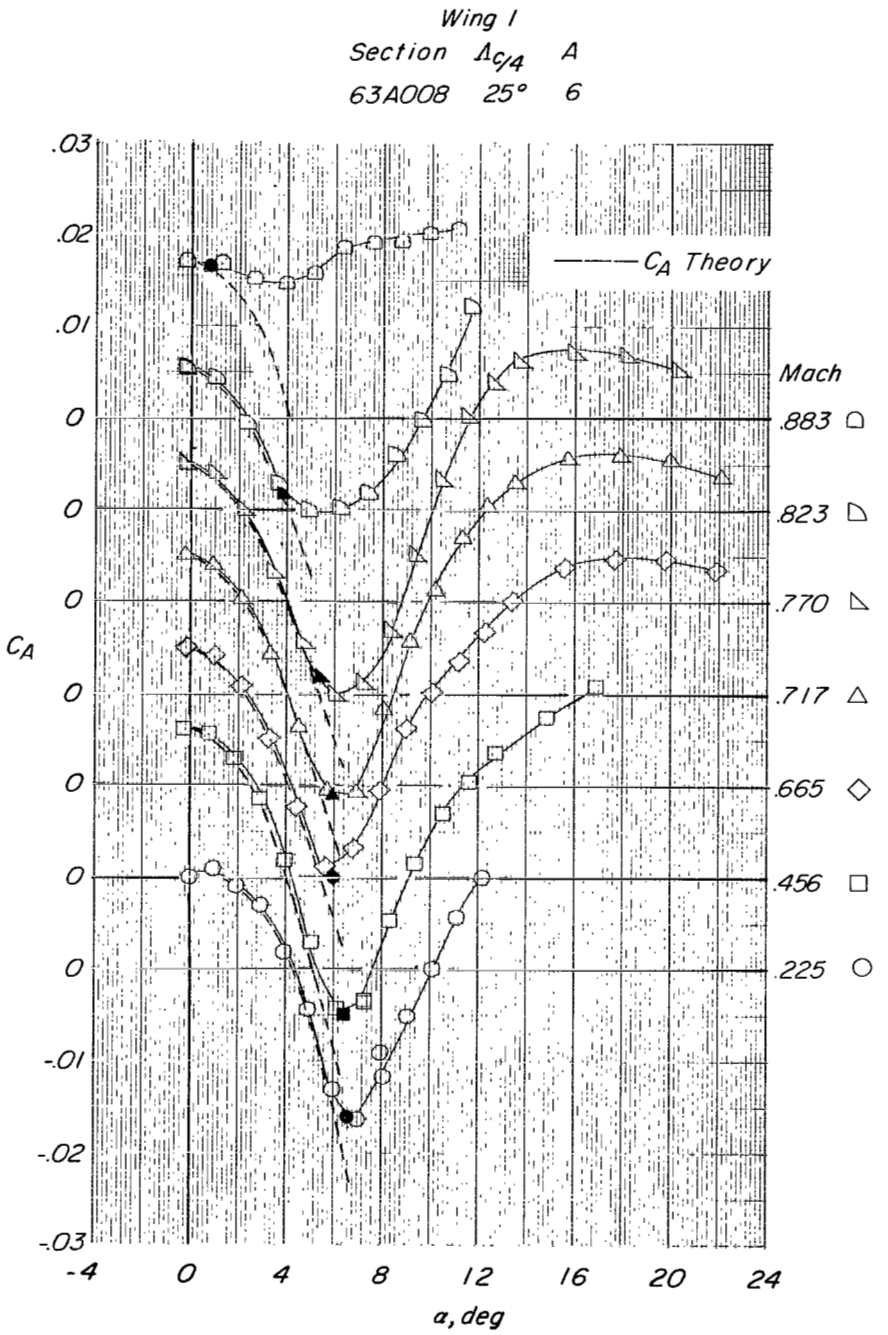
(b) Variation of dynamic pressure q with Mach number.

Figure 3.- Concluded.



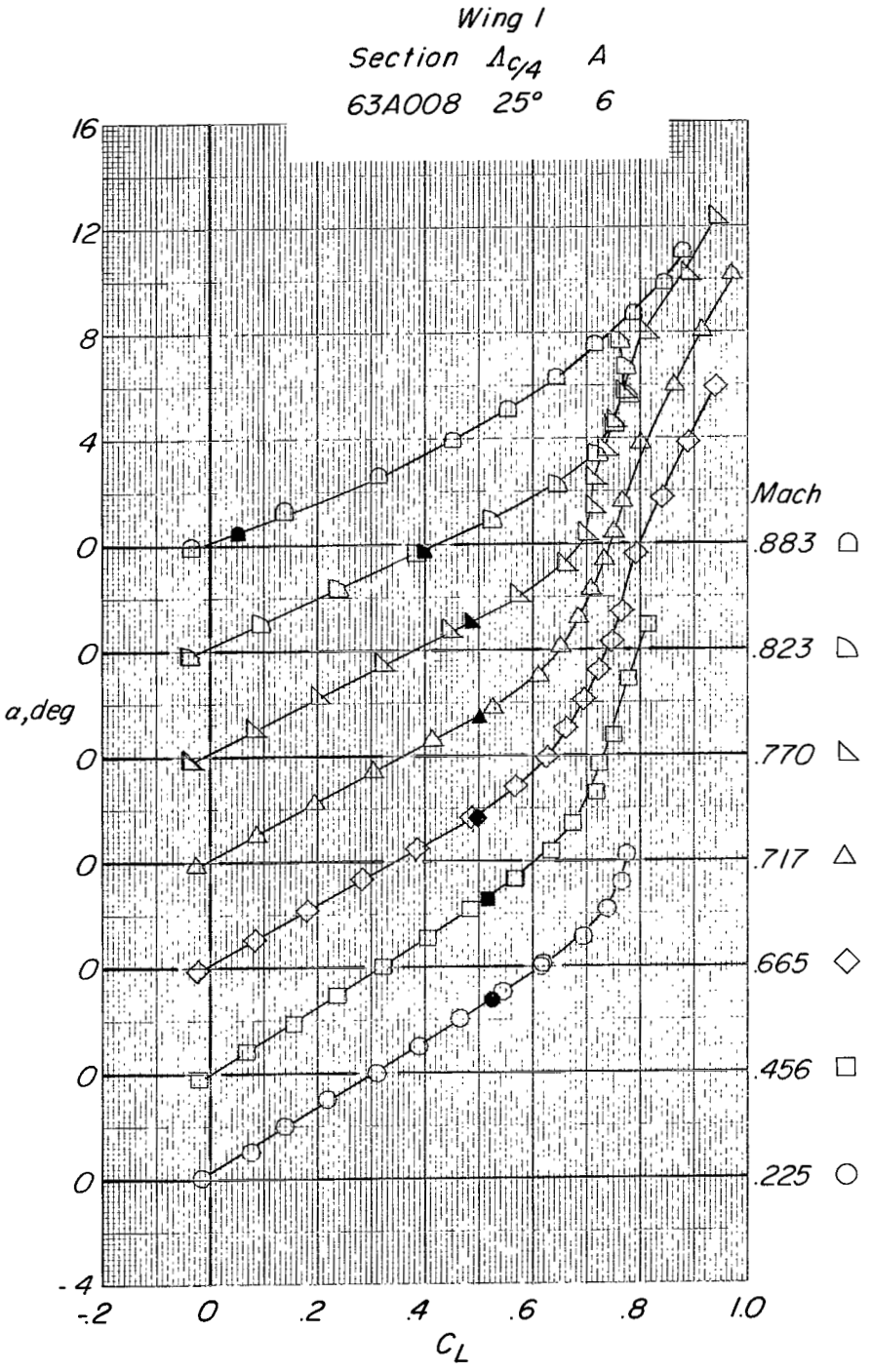
(a) C_N versus α.

Figure 4.- Static longitudinal aerodynamic and buffet characteristics of the wing 1 configuration at Mach numbers from 0.23 to 0.88. Rounded forebody; transition grit on. (Solid symbols indicate buffet onset.)



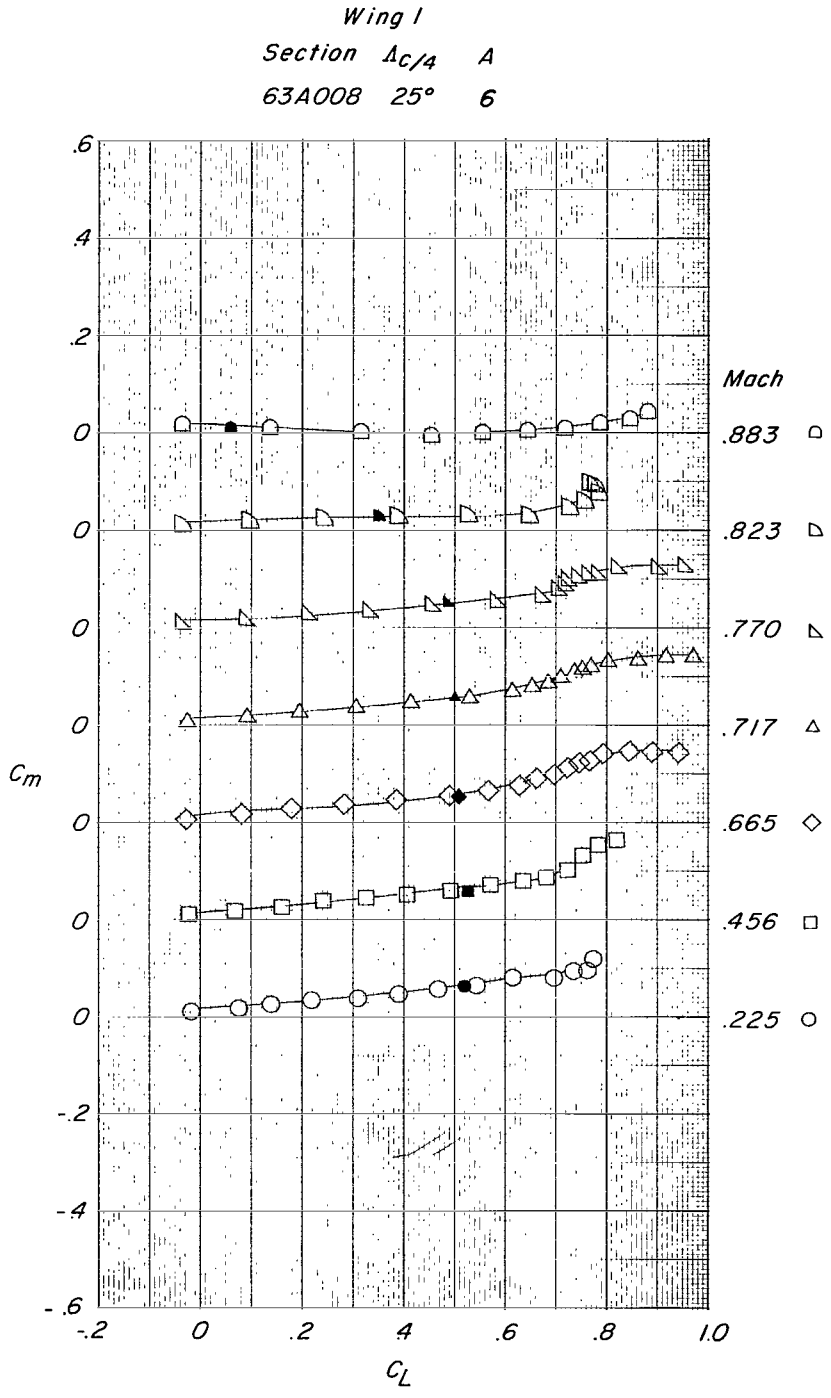
(b) C_A versus α.

Figure 4.- Continued.



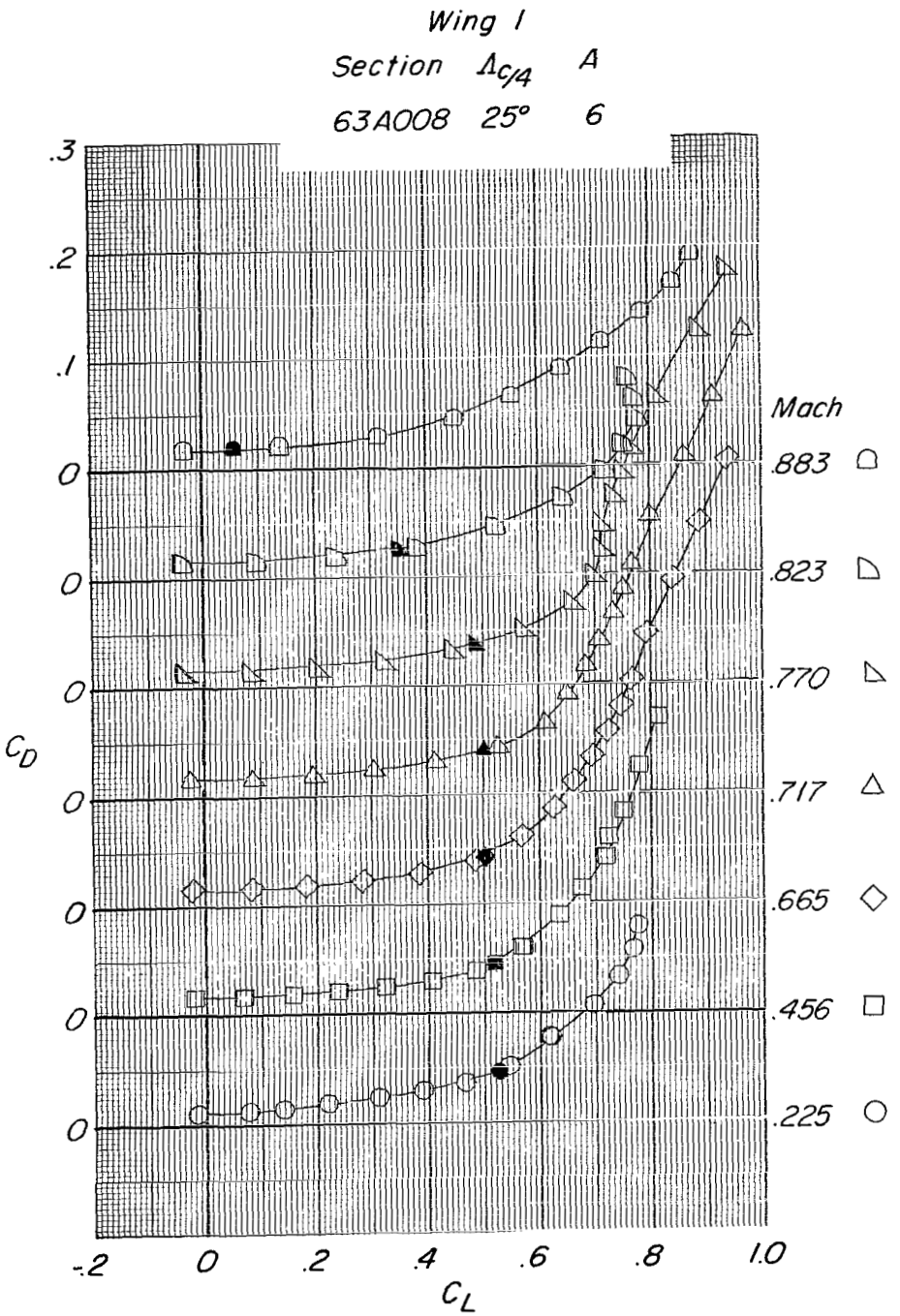
(c) α versus C_L.

Figure 4.- Continued.



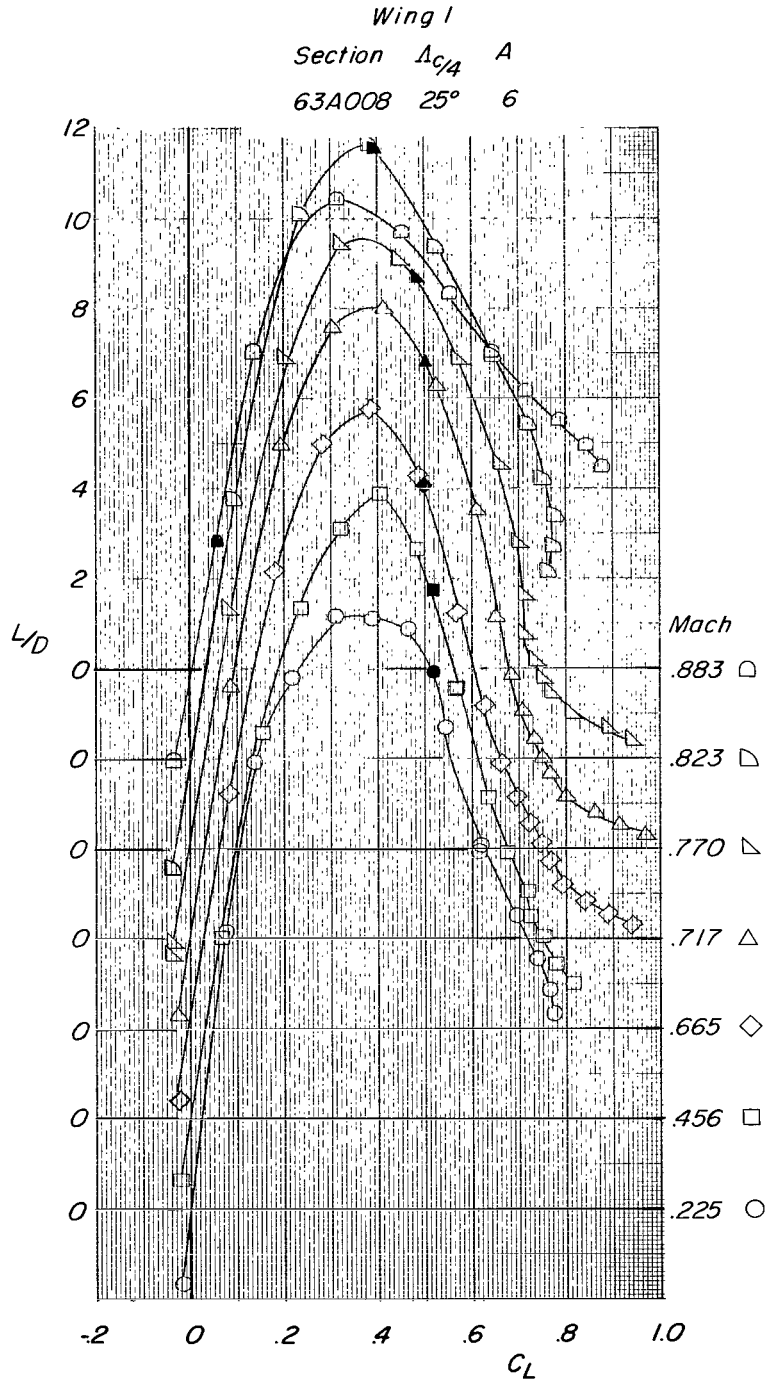
(d) C_m versus C_L.

Figure 4.- Continued.



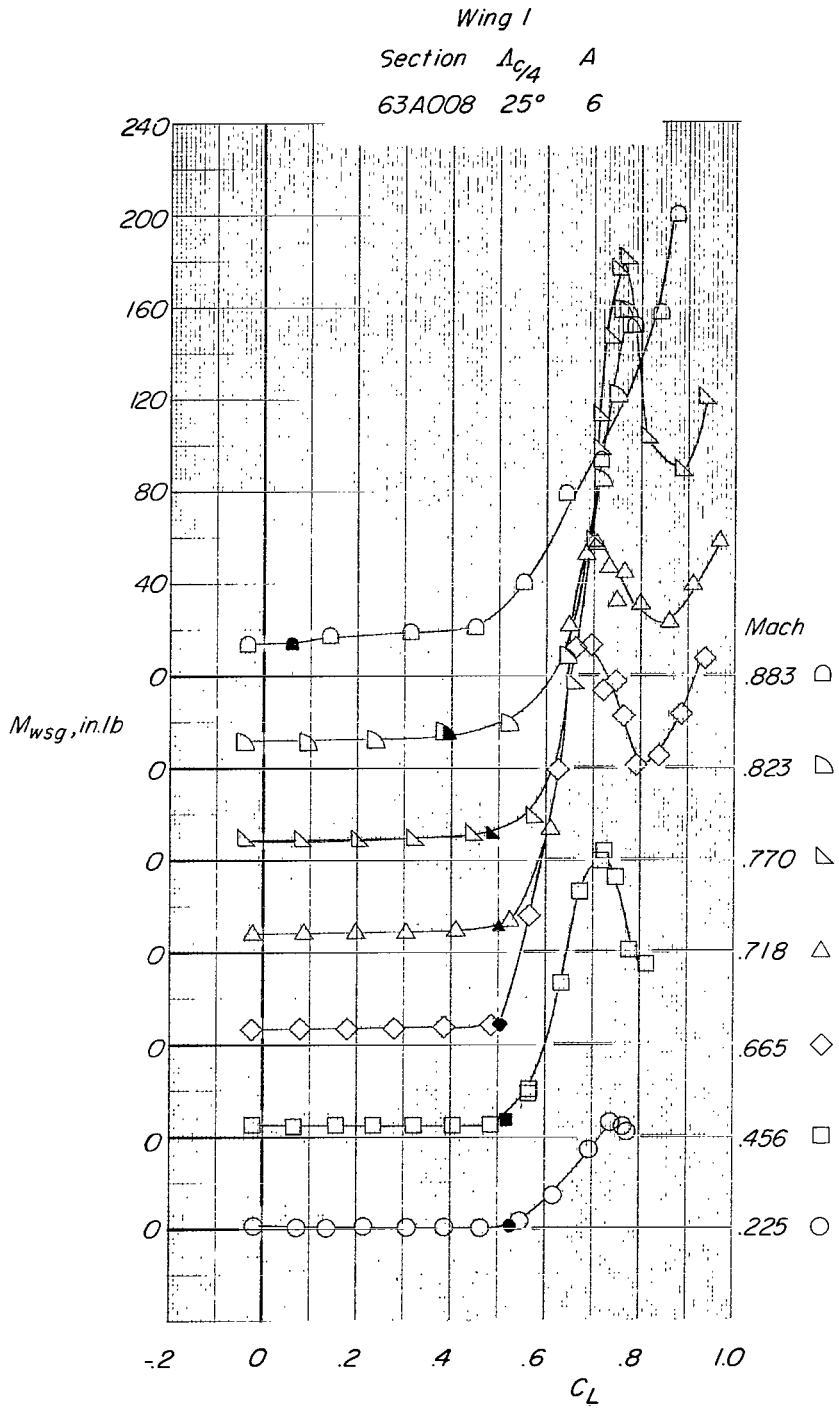
(e) C_D versus C_L.

Figure 4.- Continued.



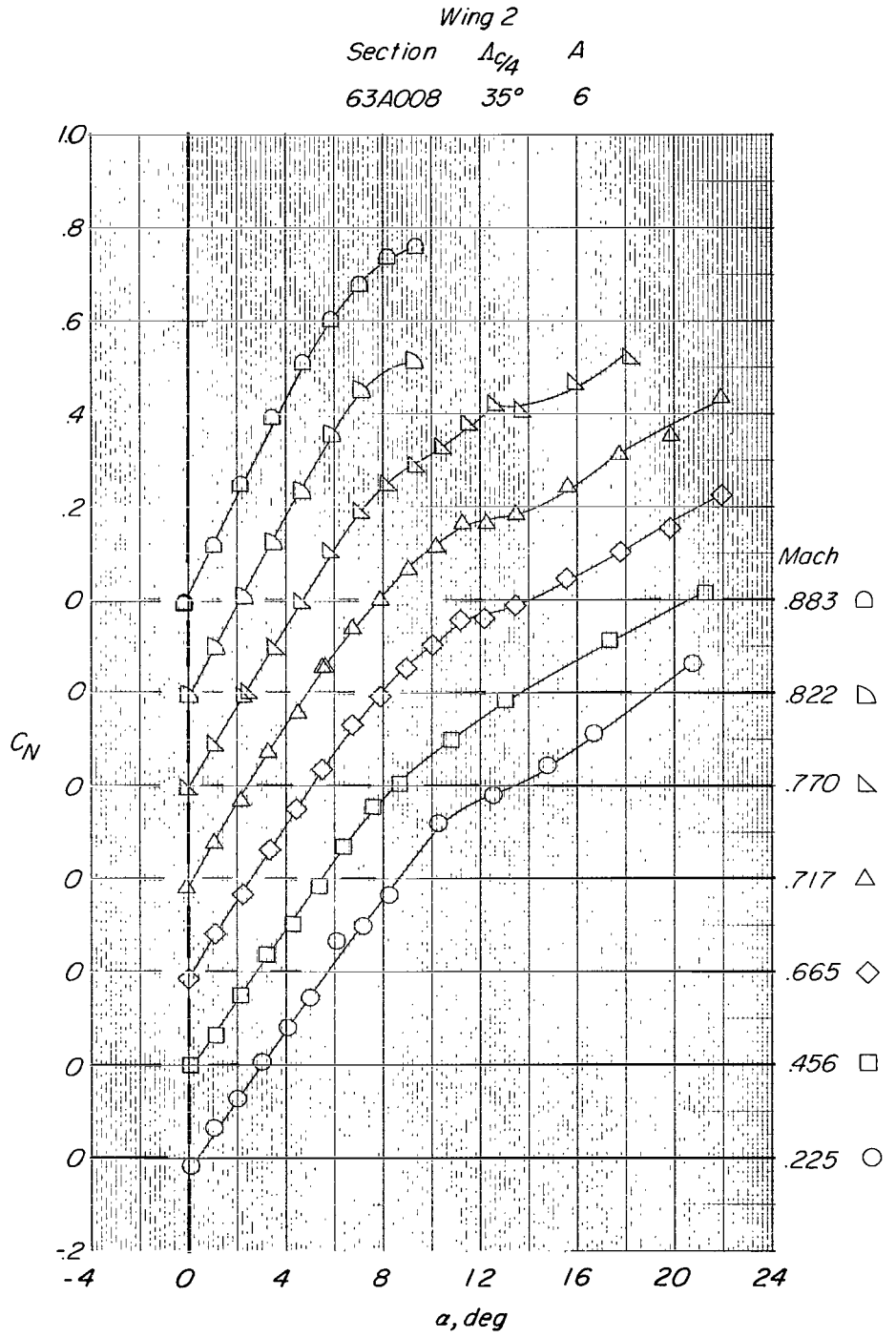
(f) L/D versus C_L.

Figure 4.- Continued.



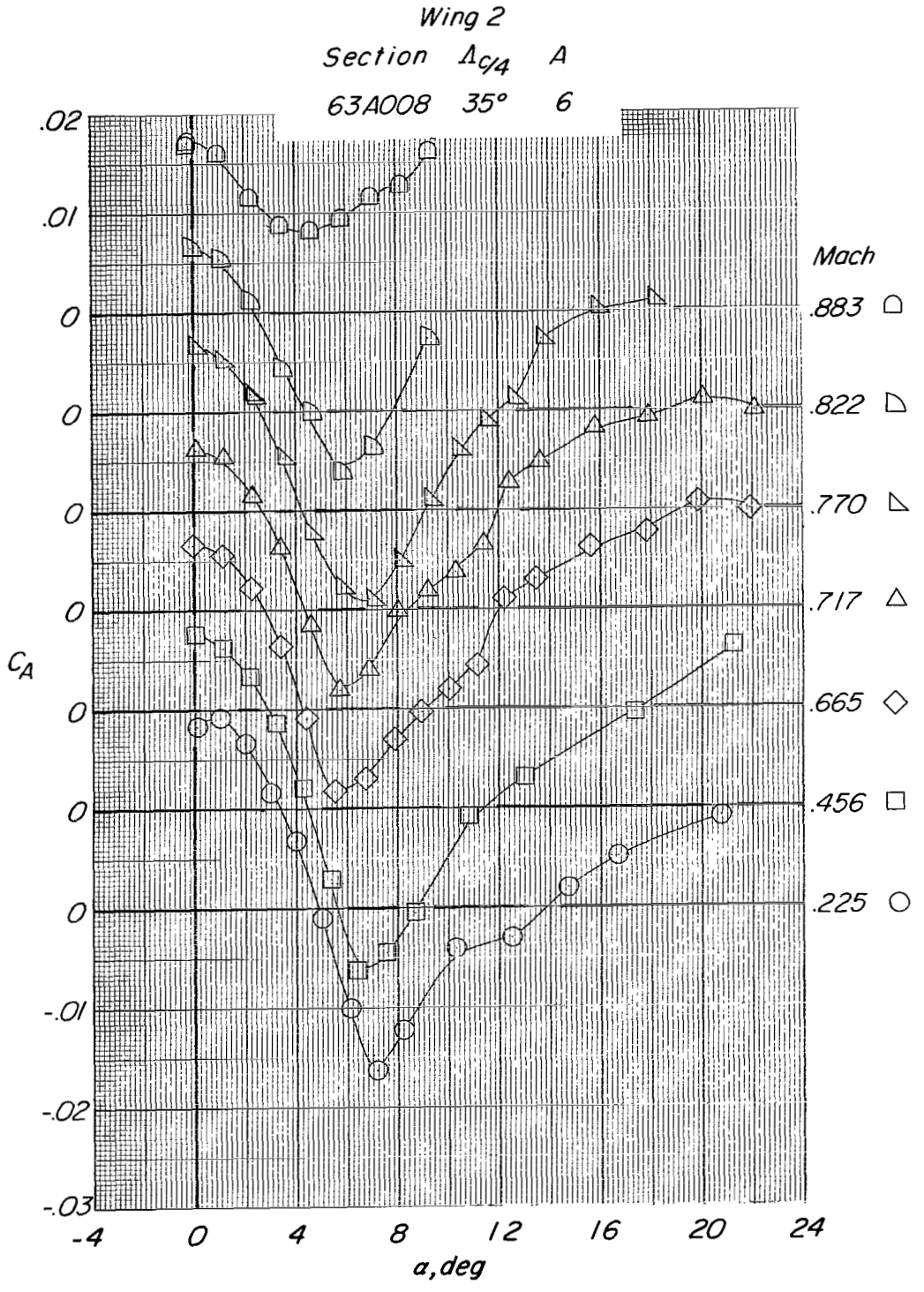
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 4.- Concluded.



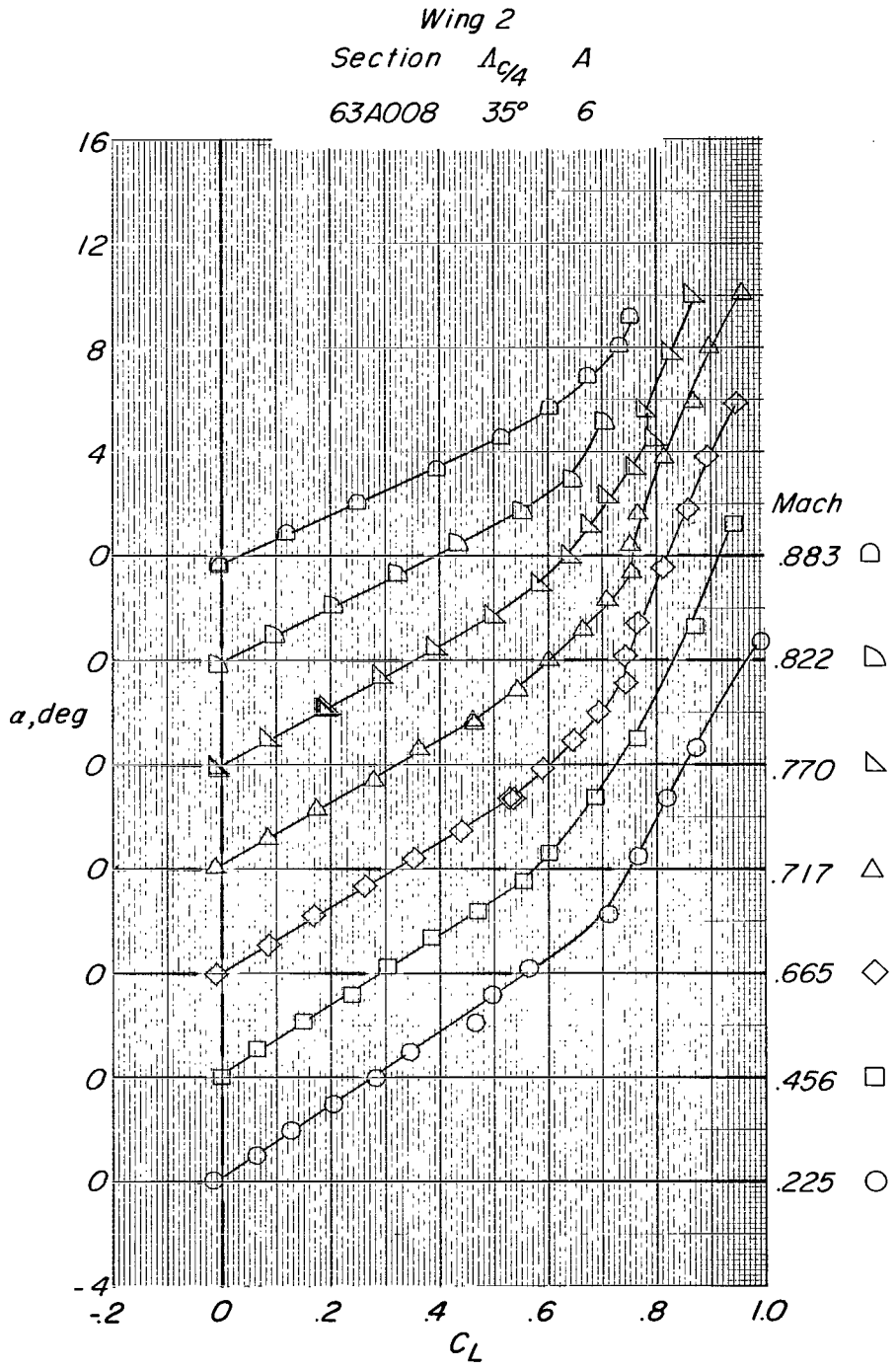
(a) C_N versus α.

Figure 5.- Static longitudinal aerodynamic and buffet characteristics of the wing 2 configuration at Mach numbers from 0.23 to 0.88. Rounded forebody; transition grit on.



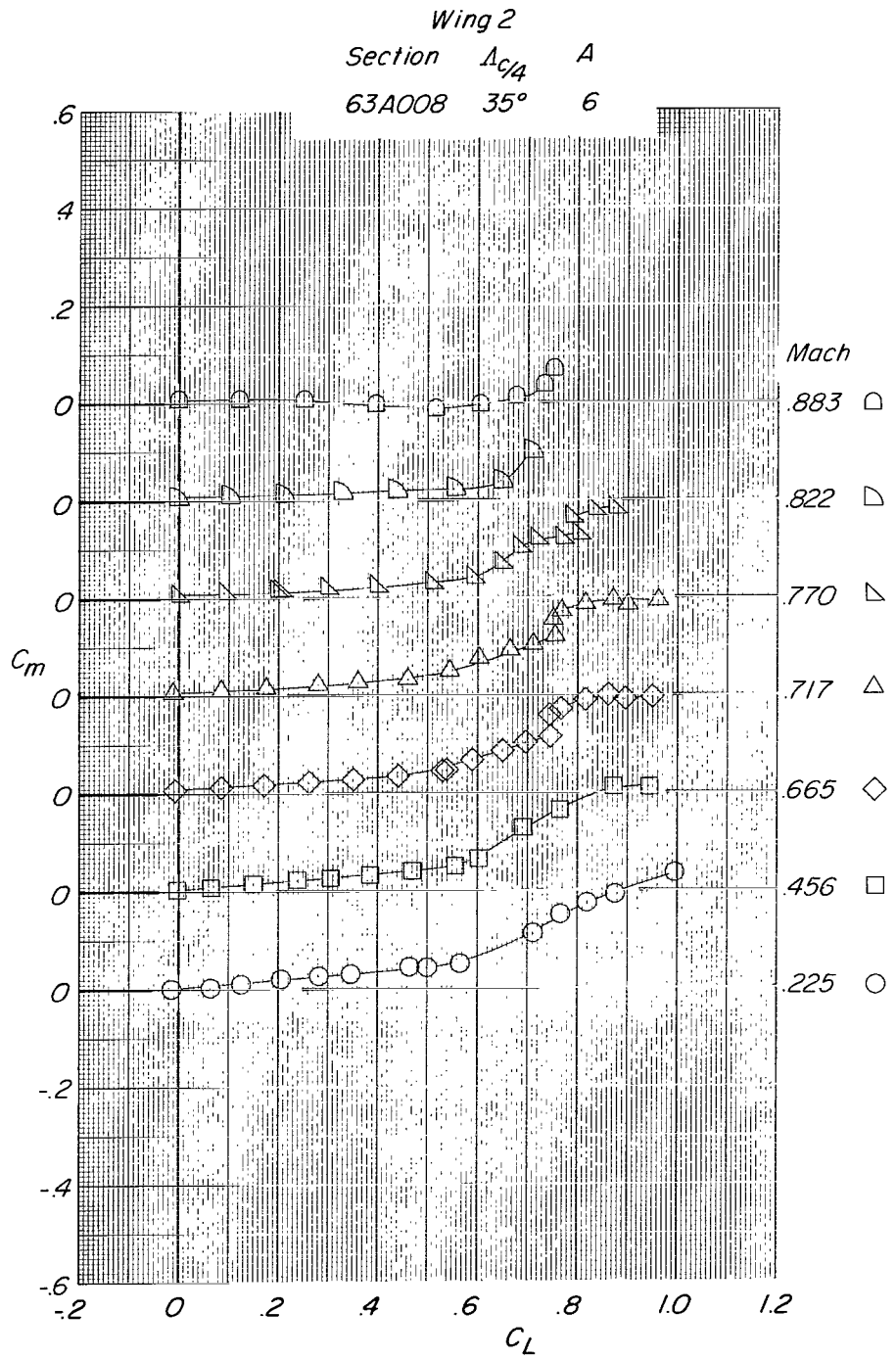
(b) C_A versus α.

Figure 5.- Continued.



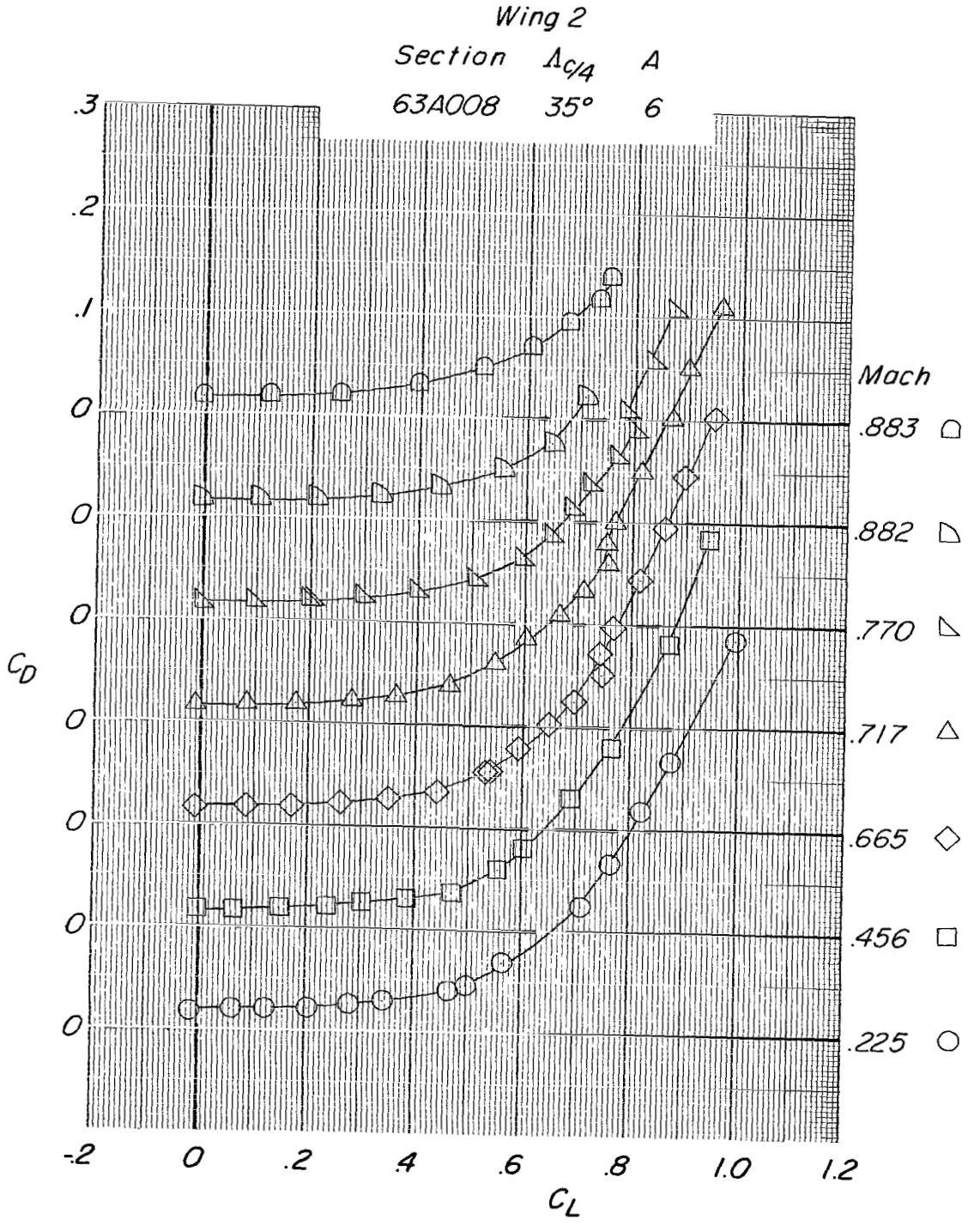
(c) α versus C_L.

Figure 5.- Continued.



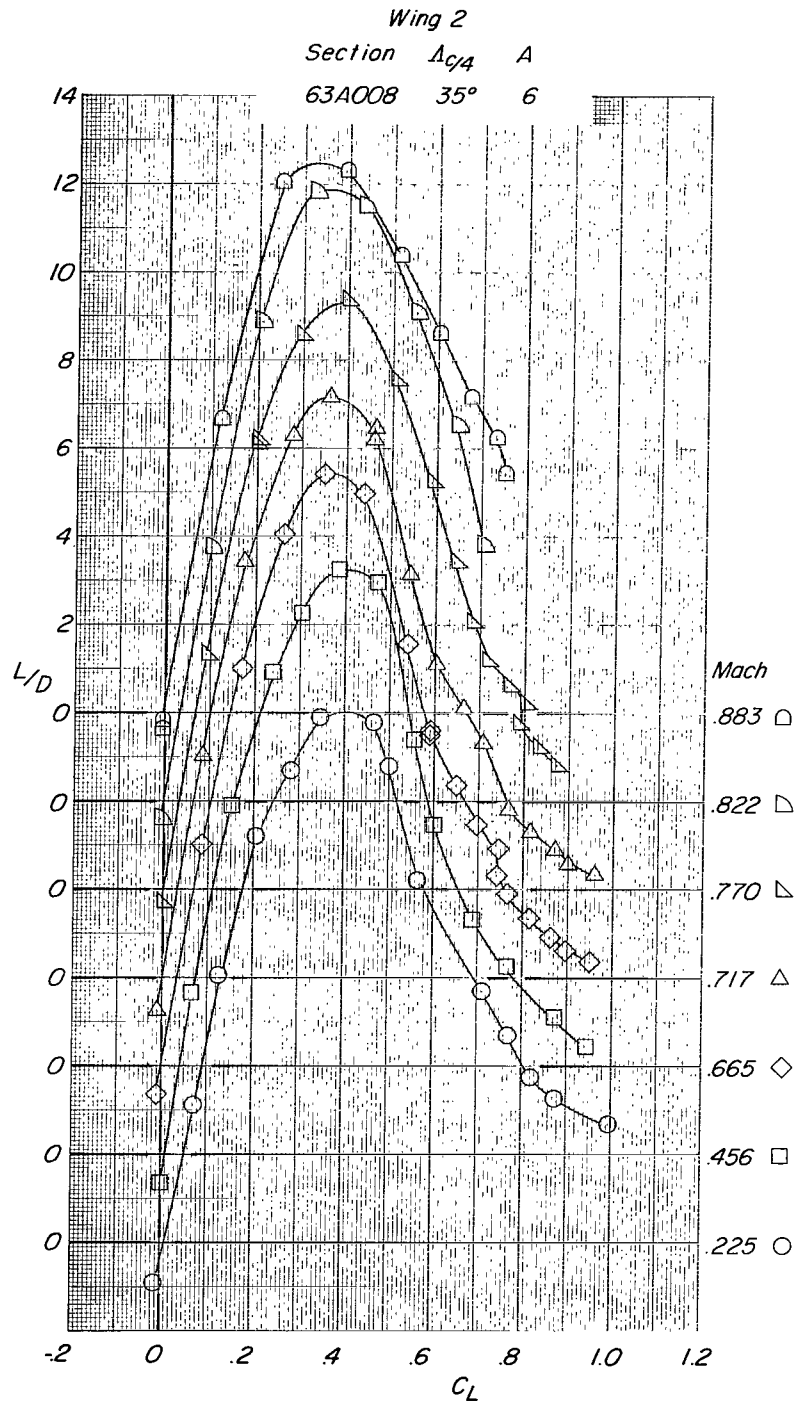
(d) C_m versus C_L.

Figure 5.- Continued.



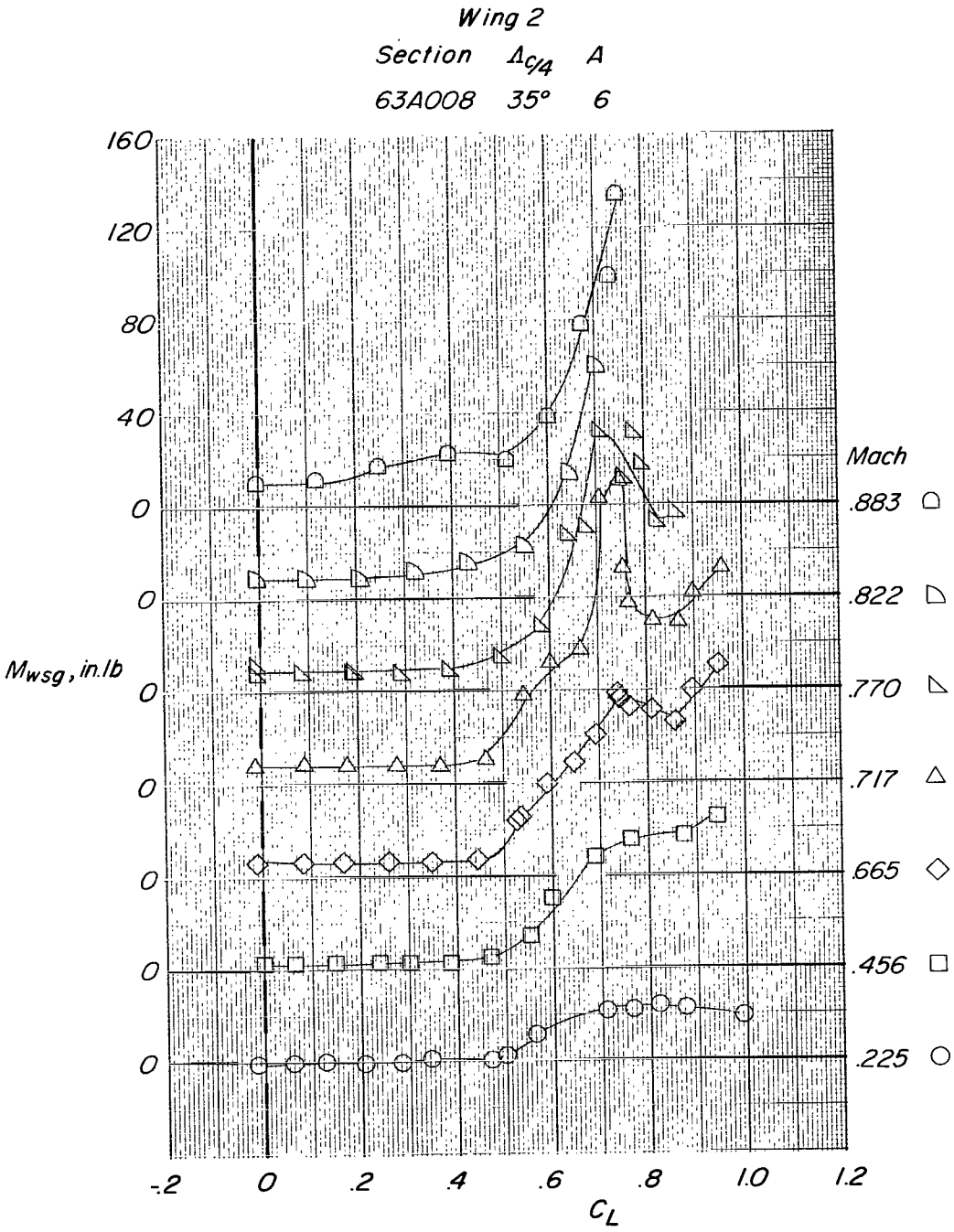
(e) C_D versus C_L.

Figure 5.- Continued.



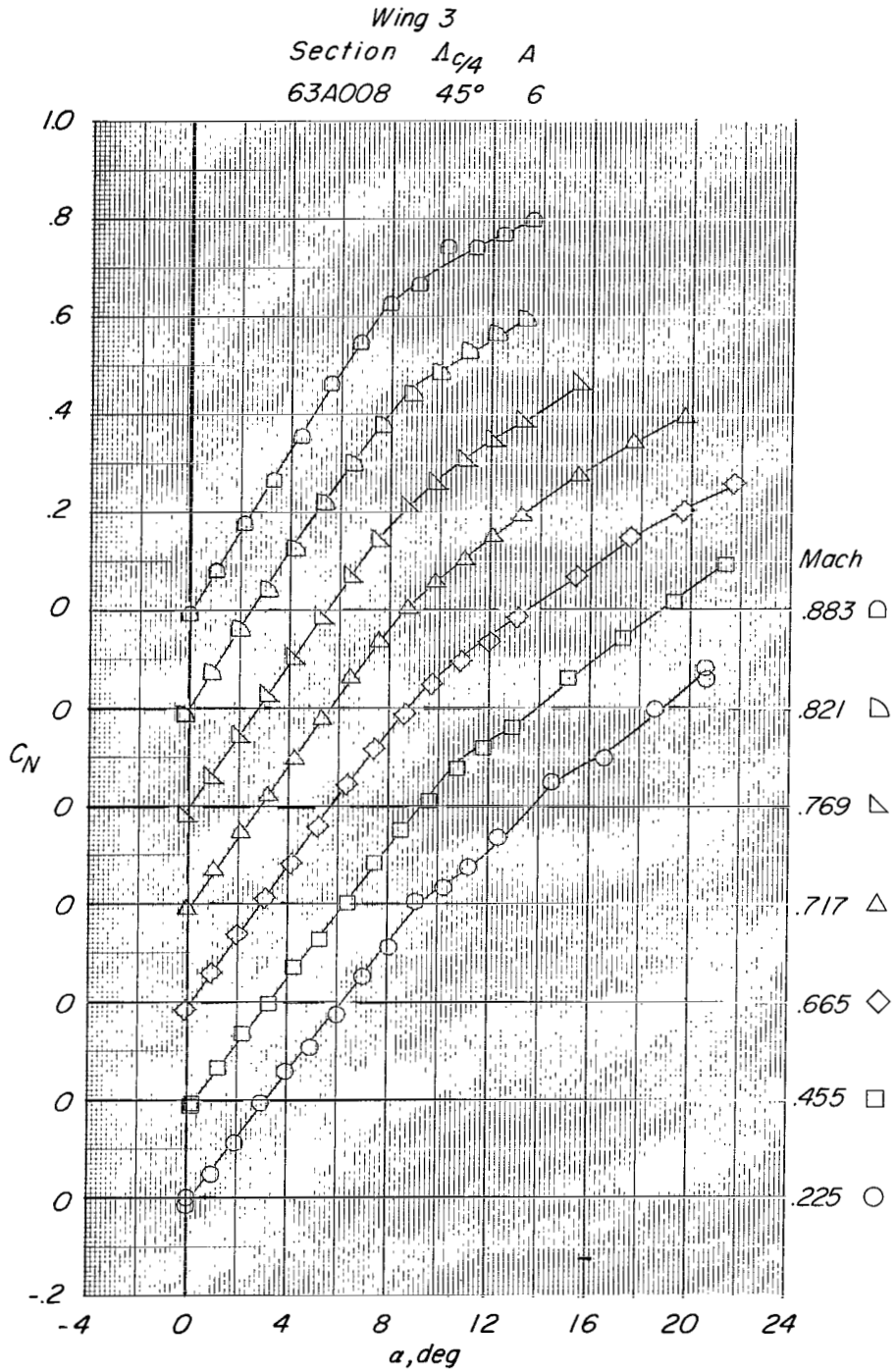
(f) L/D versus C_L.

Figure 5.- Continued.



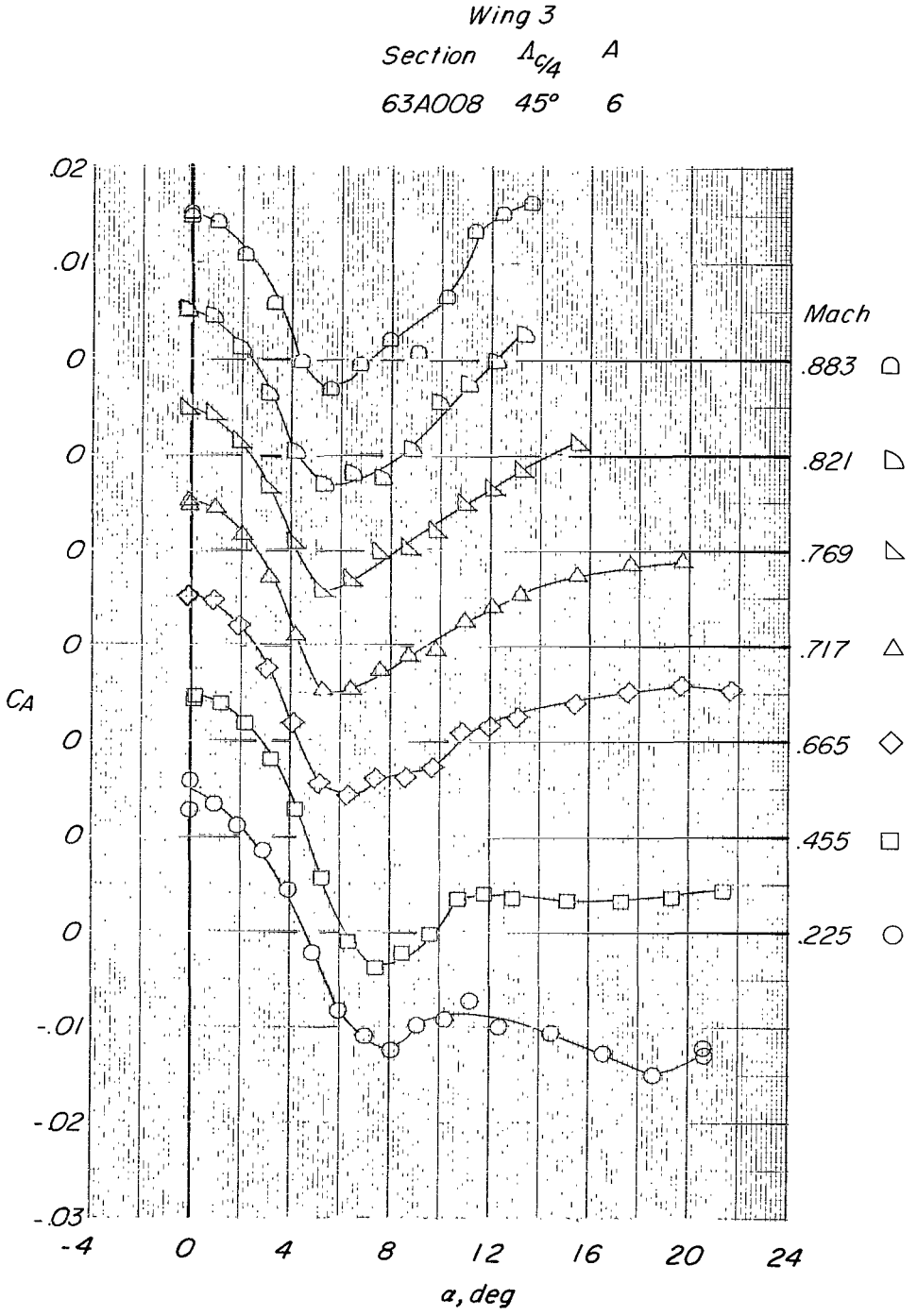
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 5.- Concluded.



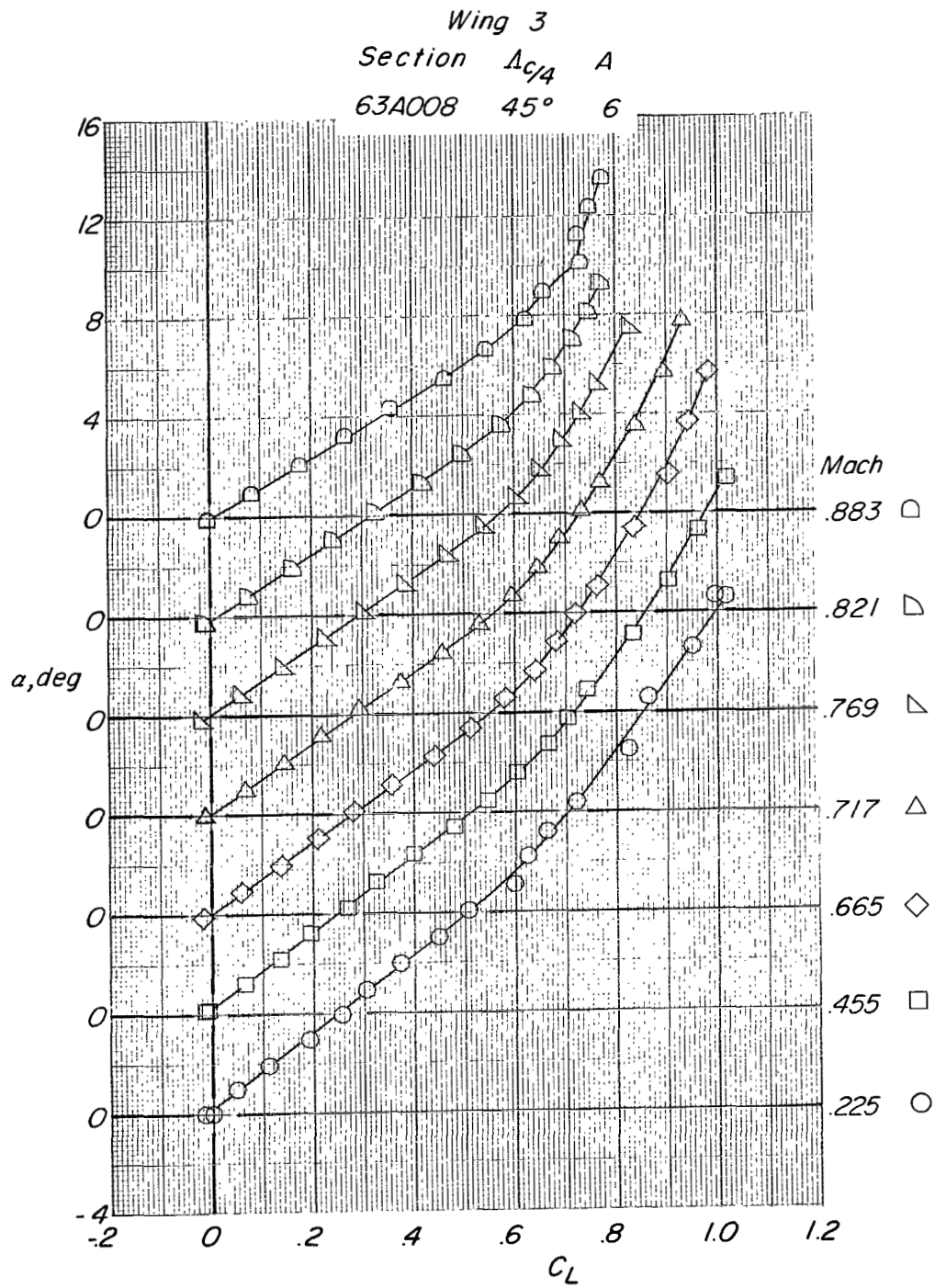
(a) C_N versus α.

Figure 6.- Static longitudinal aerodynamic and buffet characteristics of the wing 3 configuration at Mach numbers from 0.23 to 0.88. Rounded forebody; transition grit on.



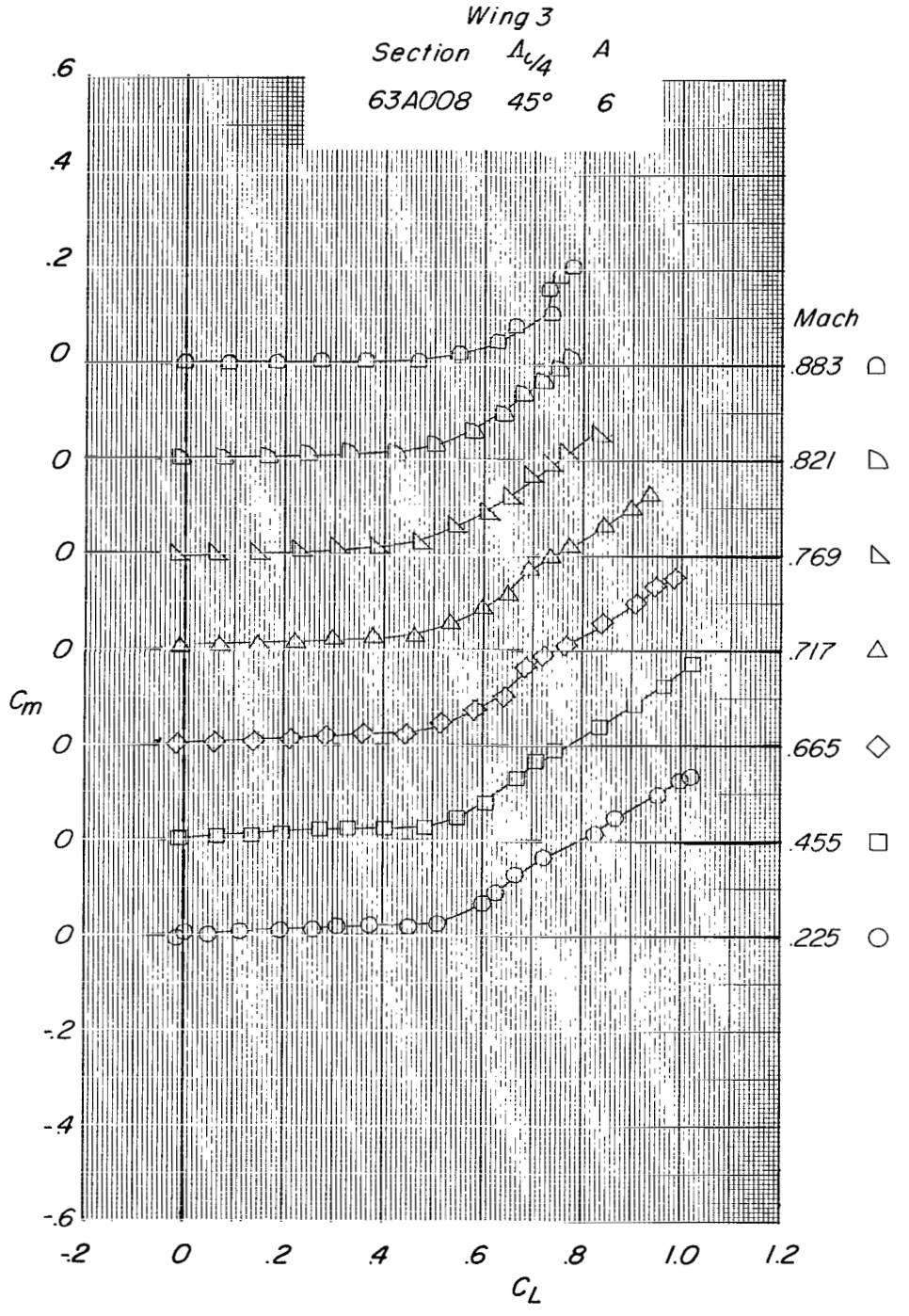
(b) C_A versus α.

Figure 6.- Continued.



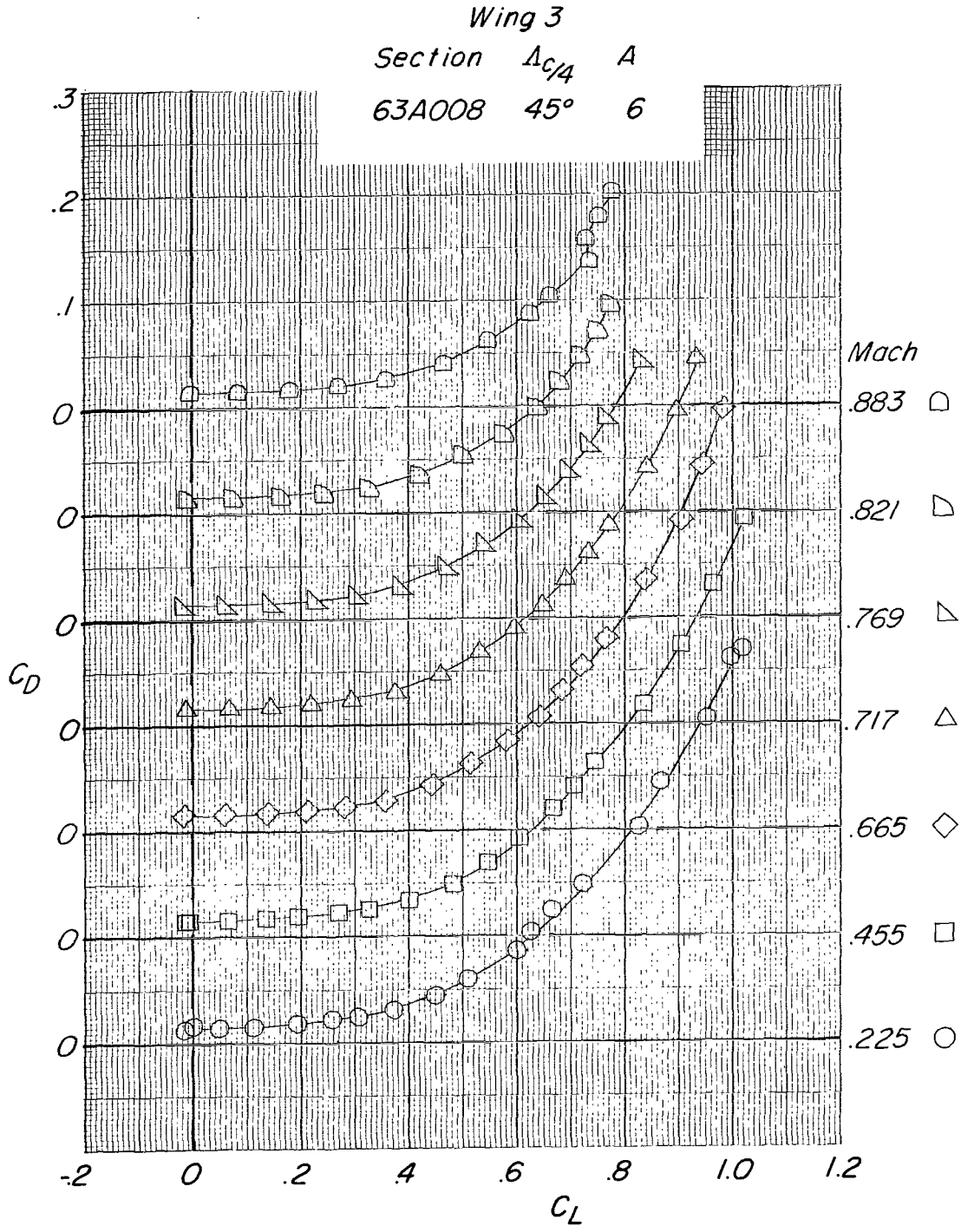
(c) α versus C_L.

Figure 6.- Continued.



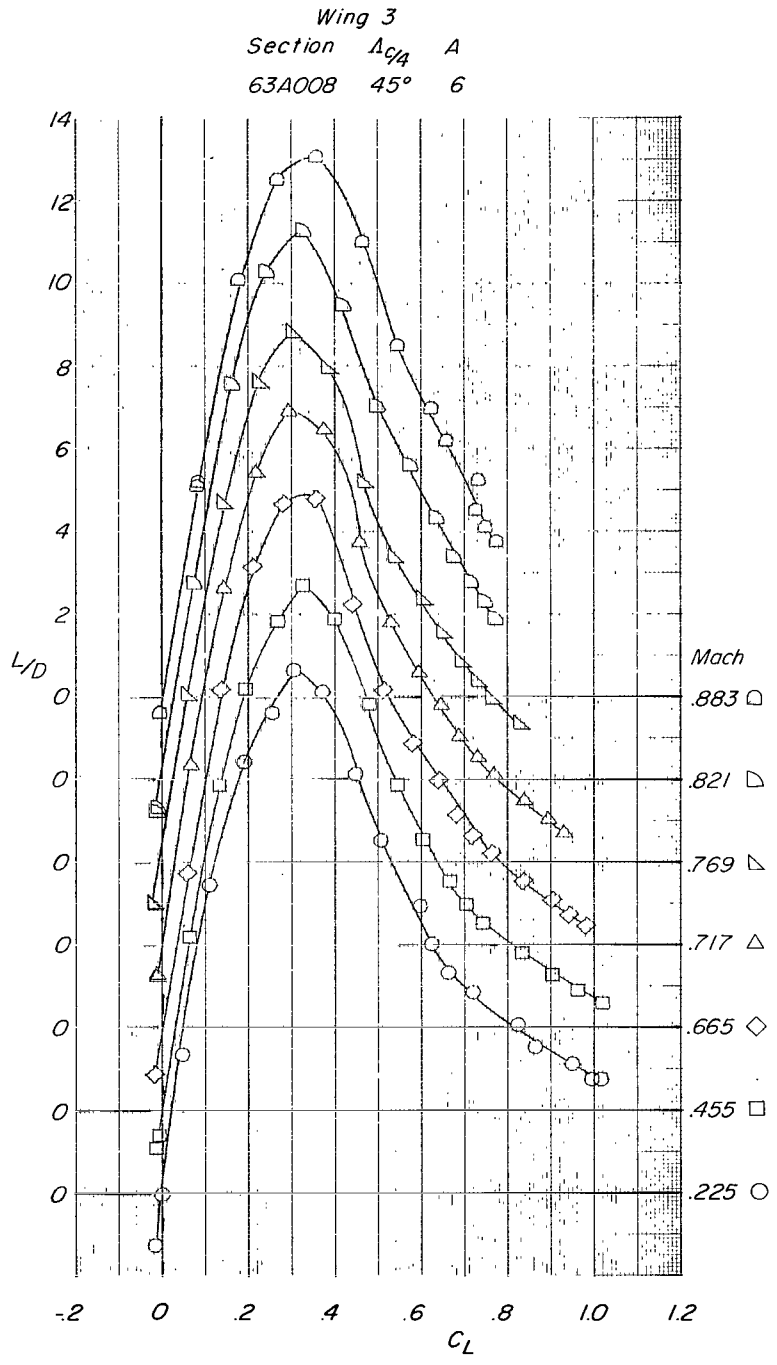
(d) C_m versus C_L.

Figure 6.- Continued.



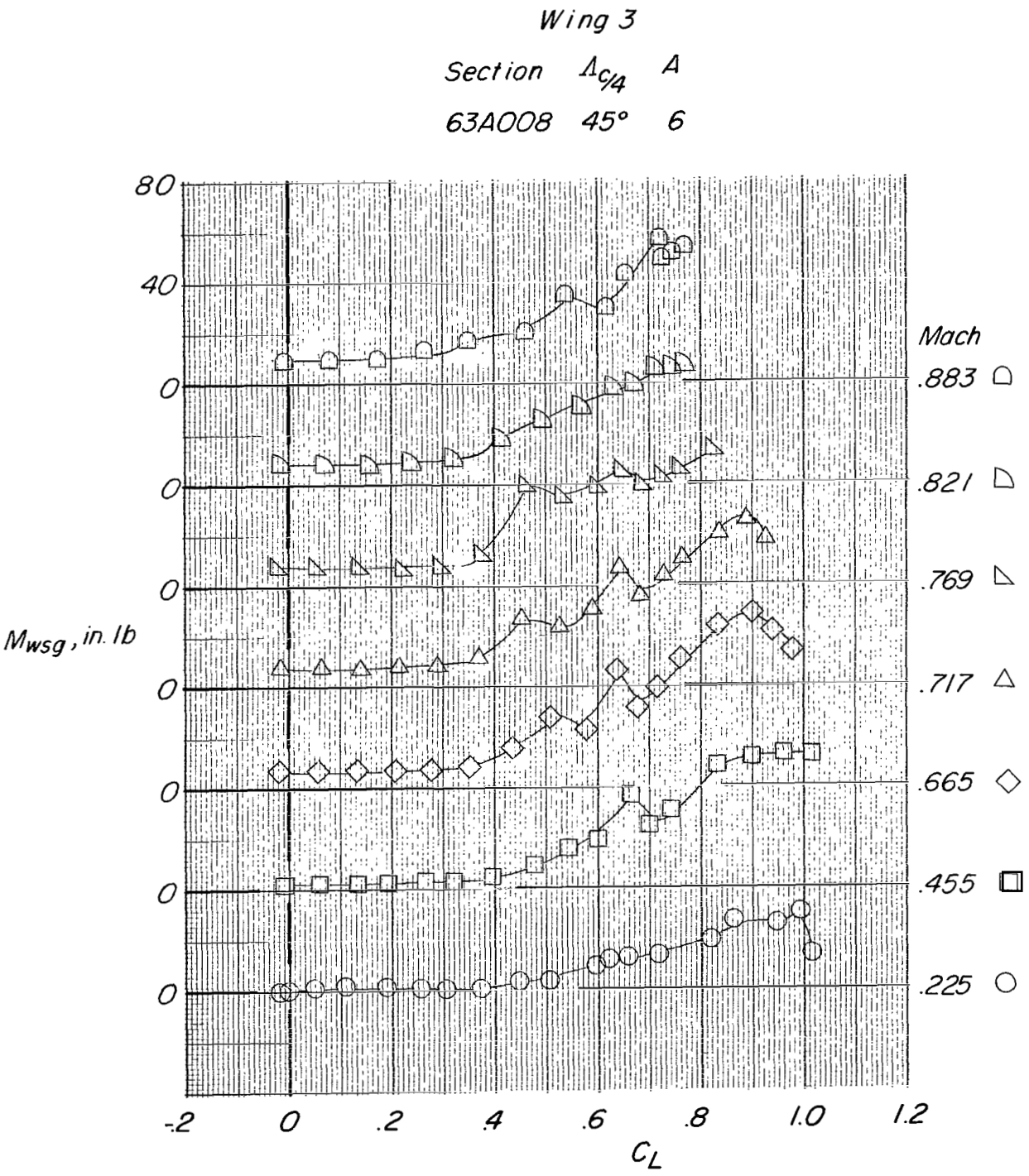
(e) C_D versus C_L.

Figure 6.- Continued.



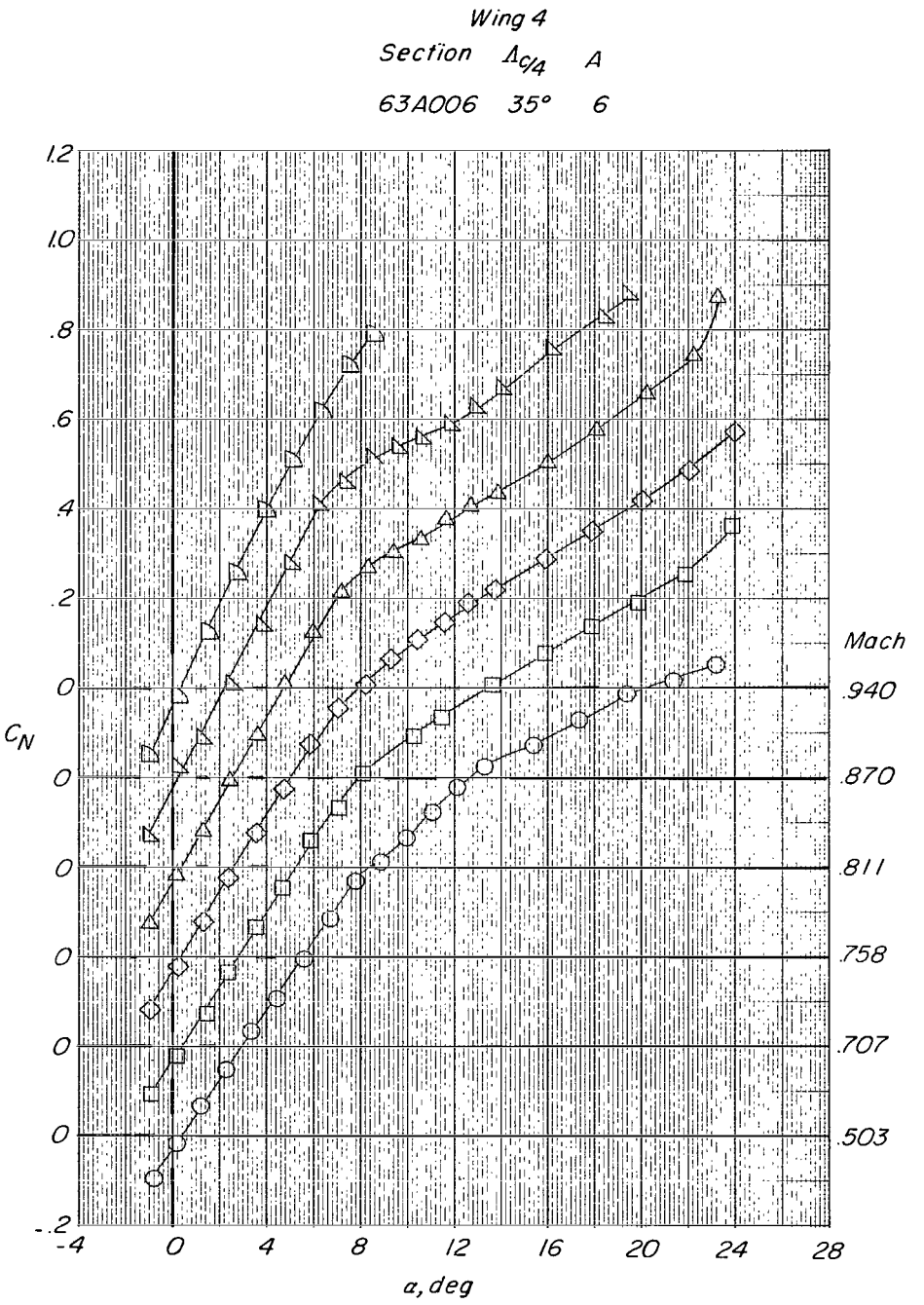
(f) L/D versus C_L.

Figure 6.- Continued.



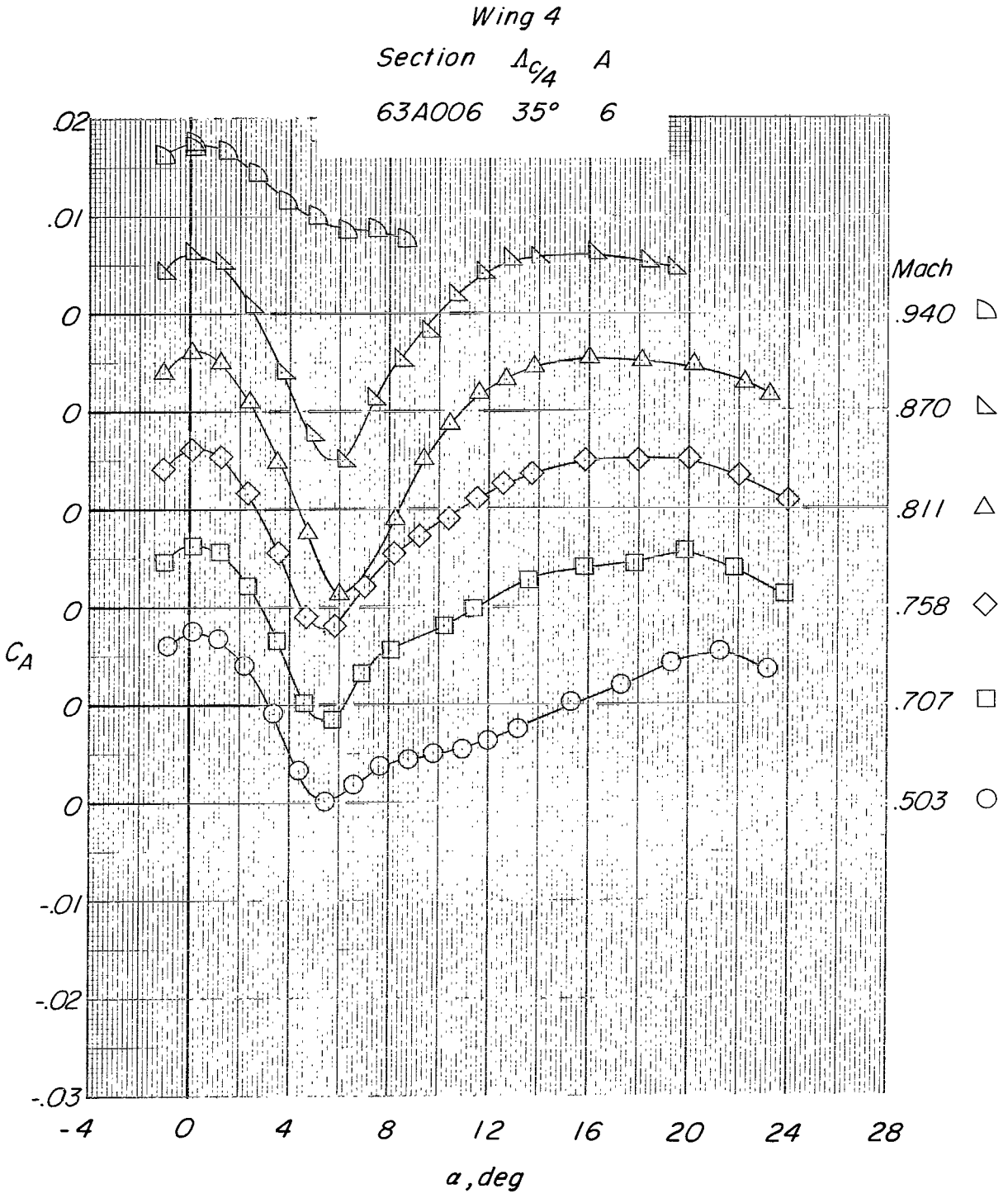
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 6.- Concluded.



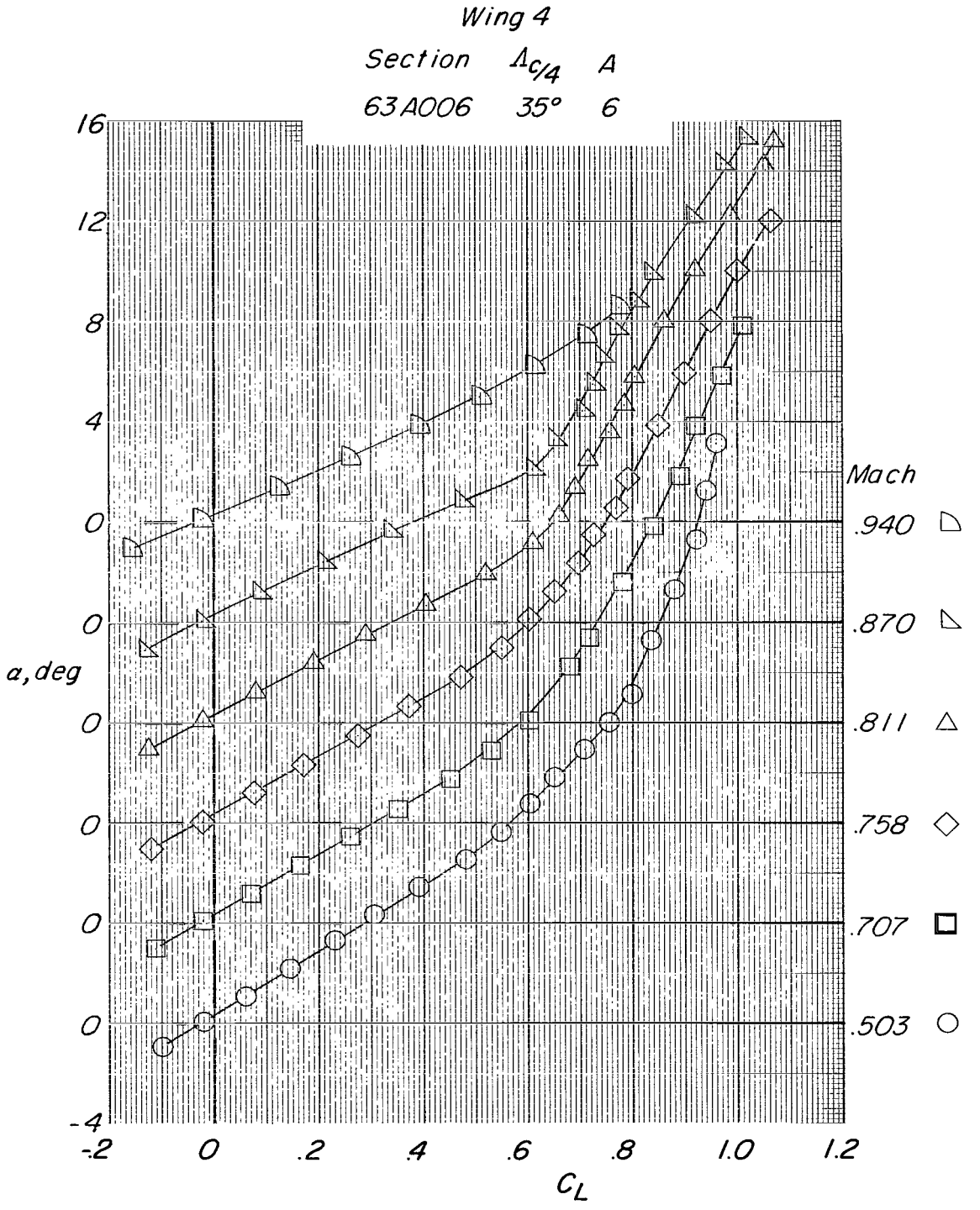
(a) C_N versus α.

Figure 7.- Static longitudinal aerodynamic and buffet characteristics of the wing 4 configuration at Mach numbers from 0.50 to 0.94. Rounded forebody; transition grit on.



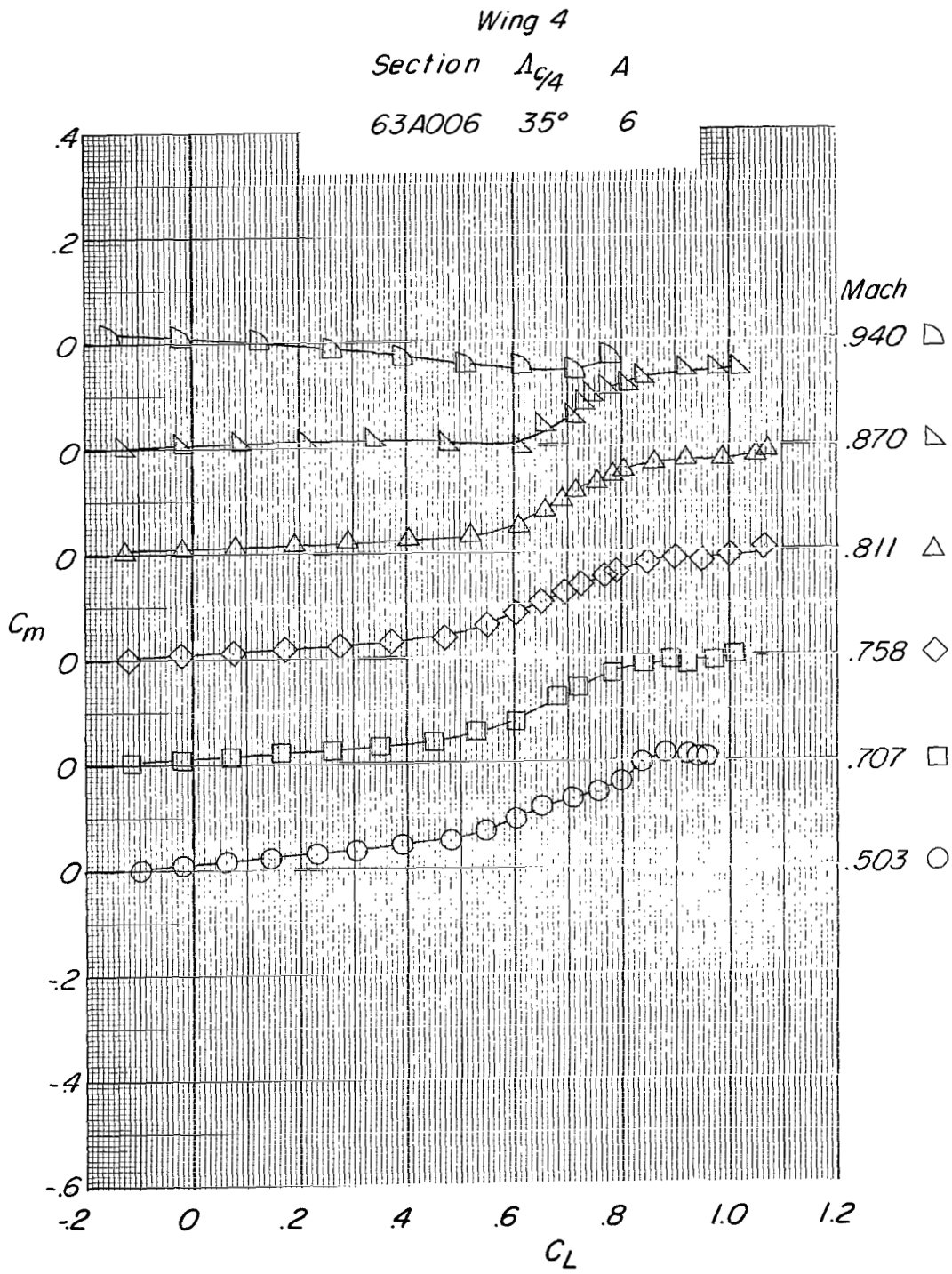
(b) C_A versus α.

Figure 7.- Continued.



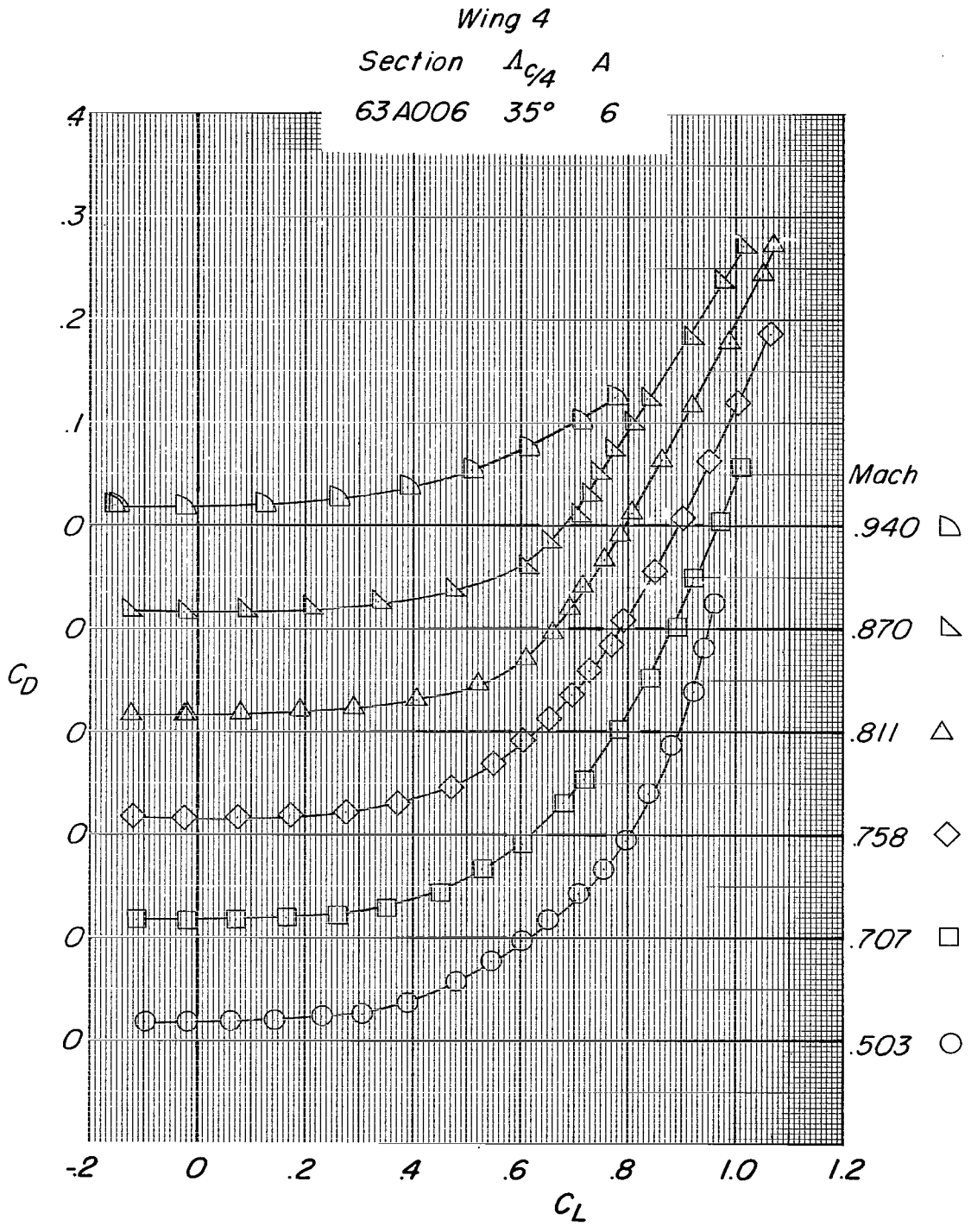
(c) α versus C_L.

Figure 7.- Continued.



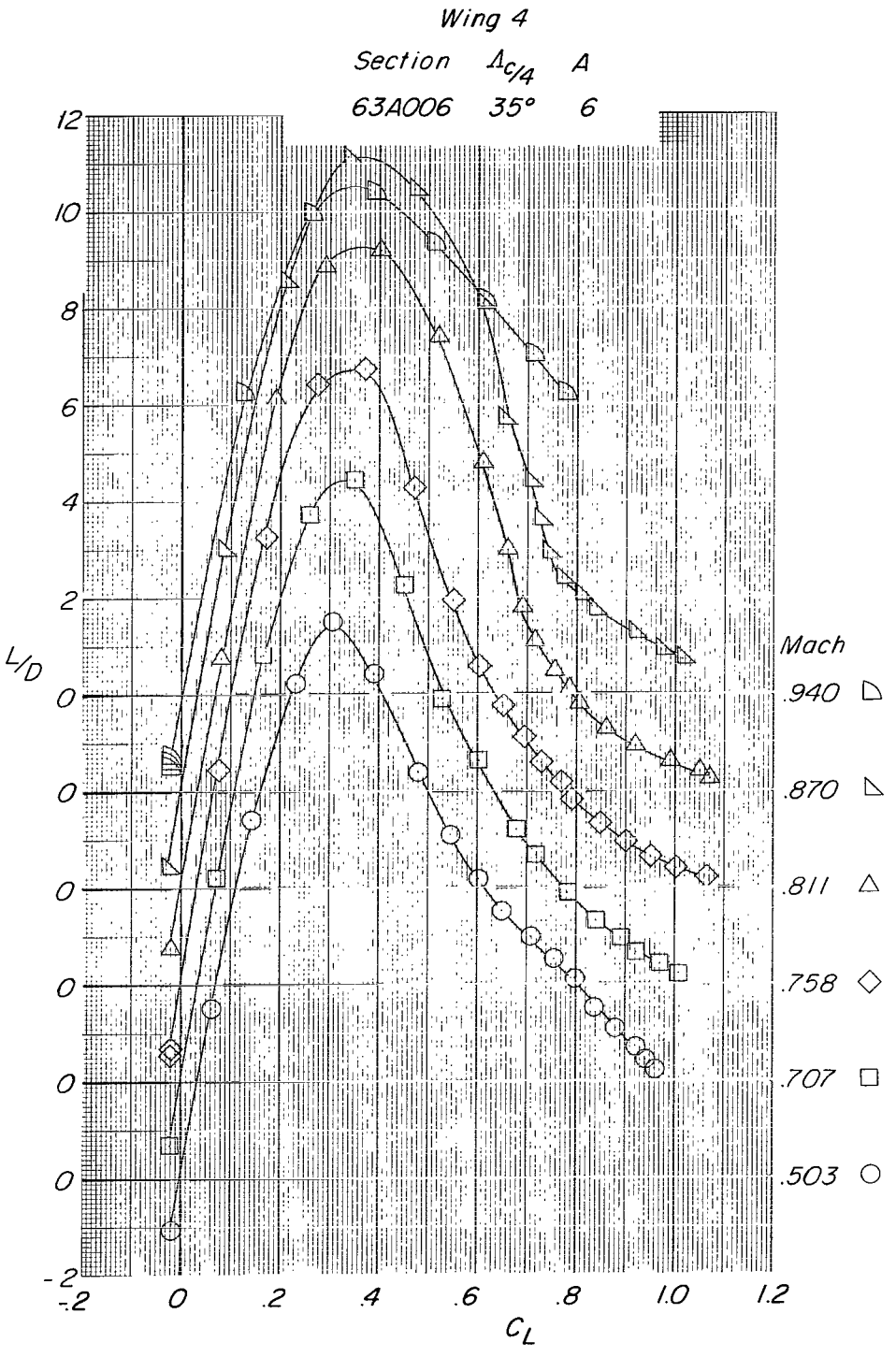
(d) C_m versus C_L.

Figure 7.- Continued.



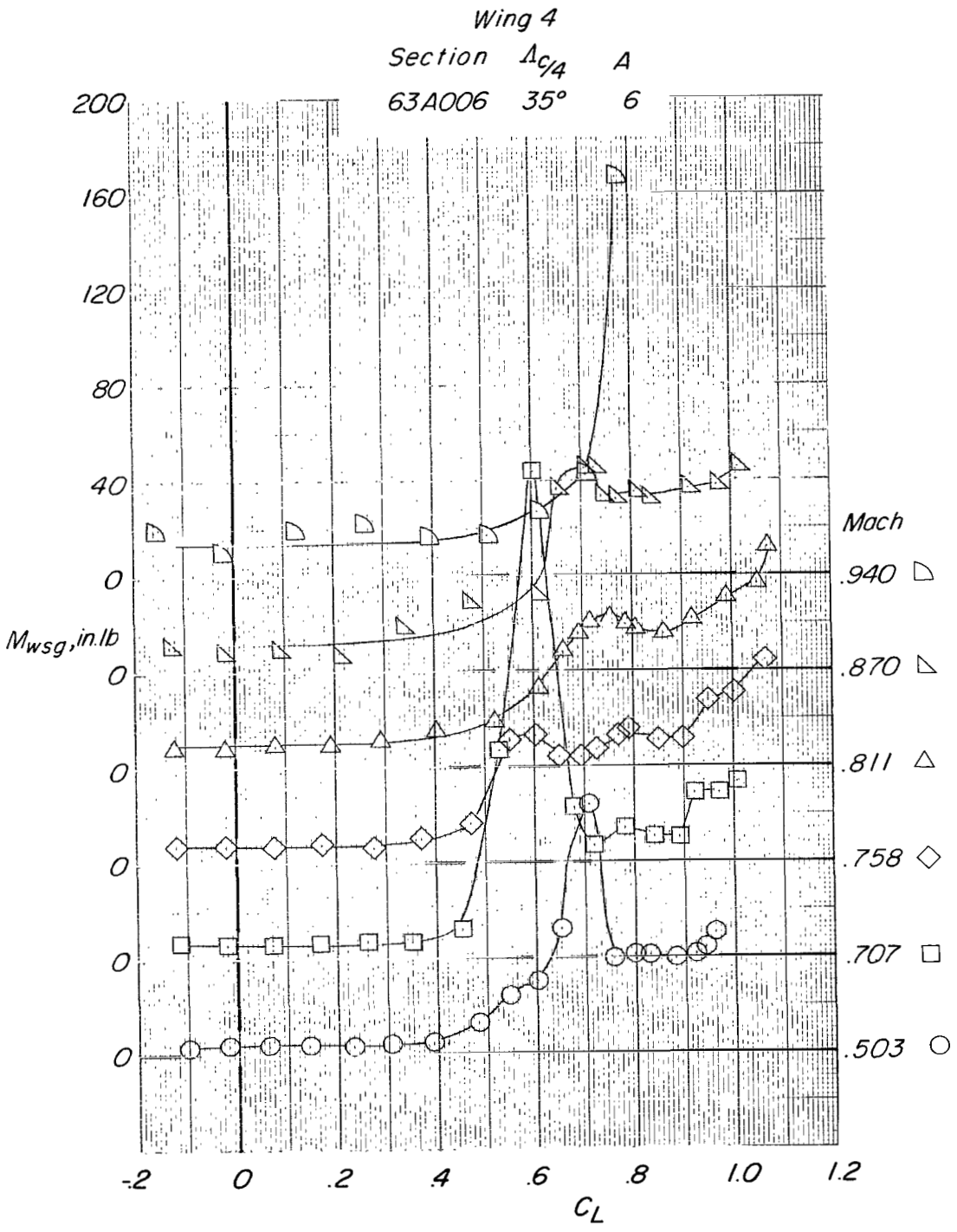
(e) C_D versus C_L.

Figure 7.- Continued.



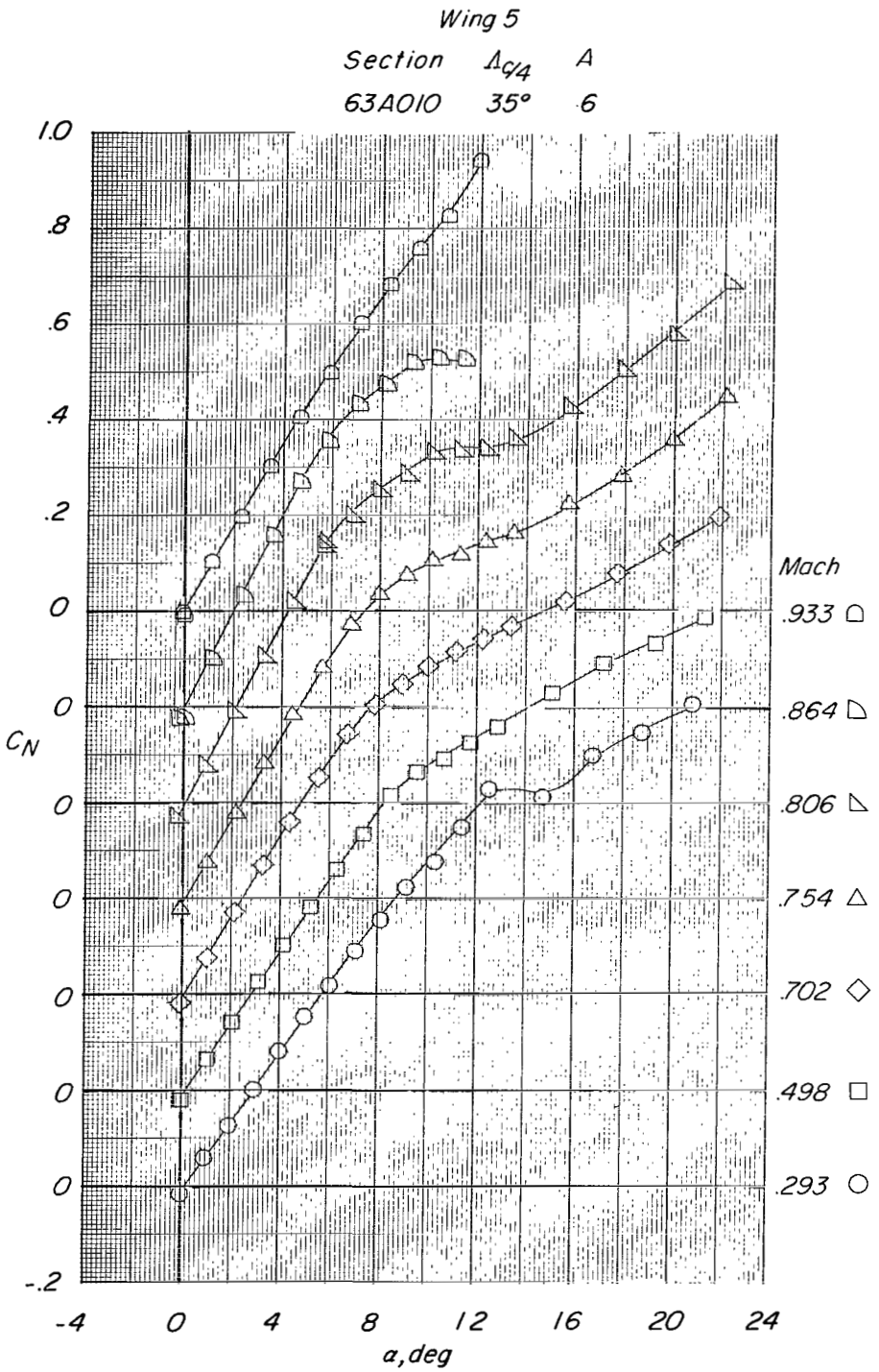
(f) L/D versus C_L.

Figure 7.- Continued.



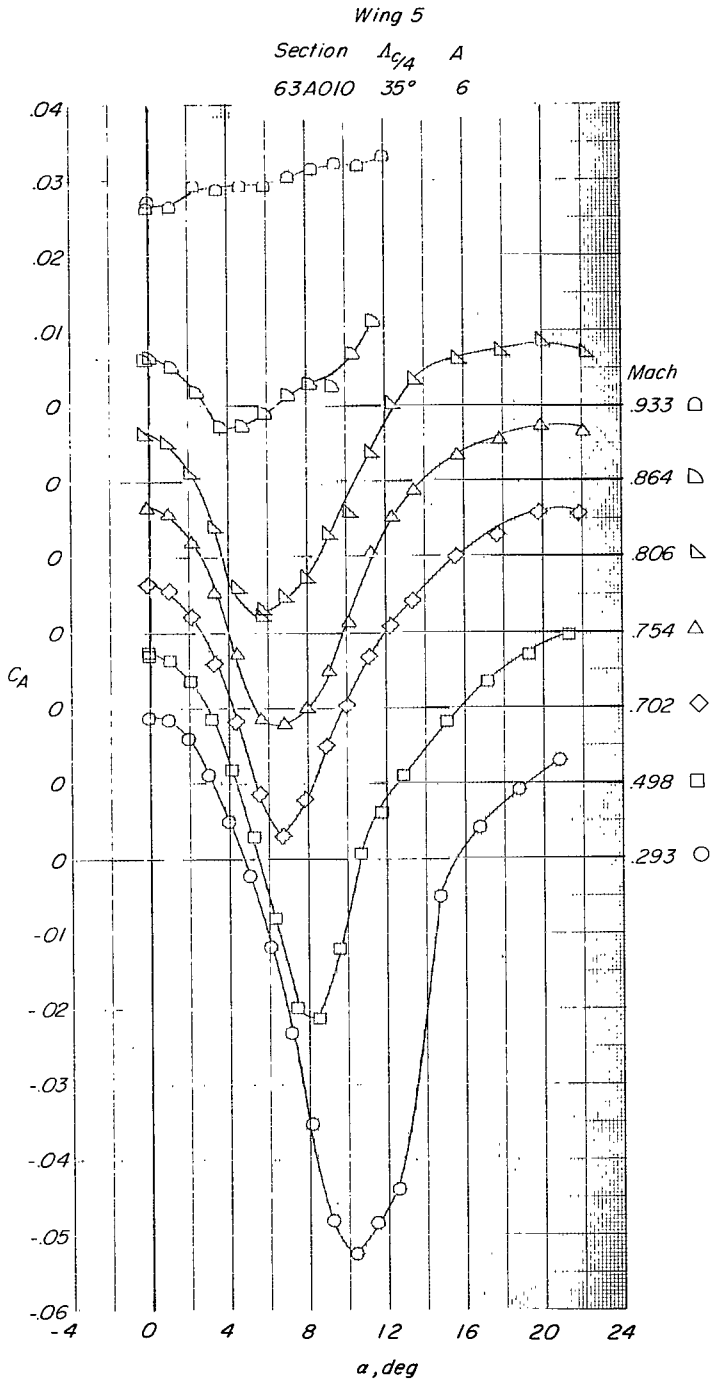
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 7.- Concluded.



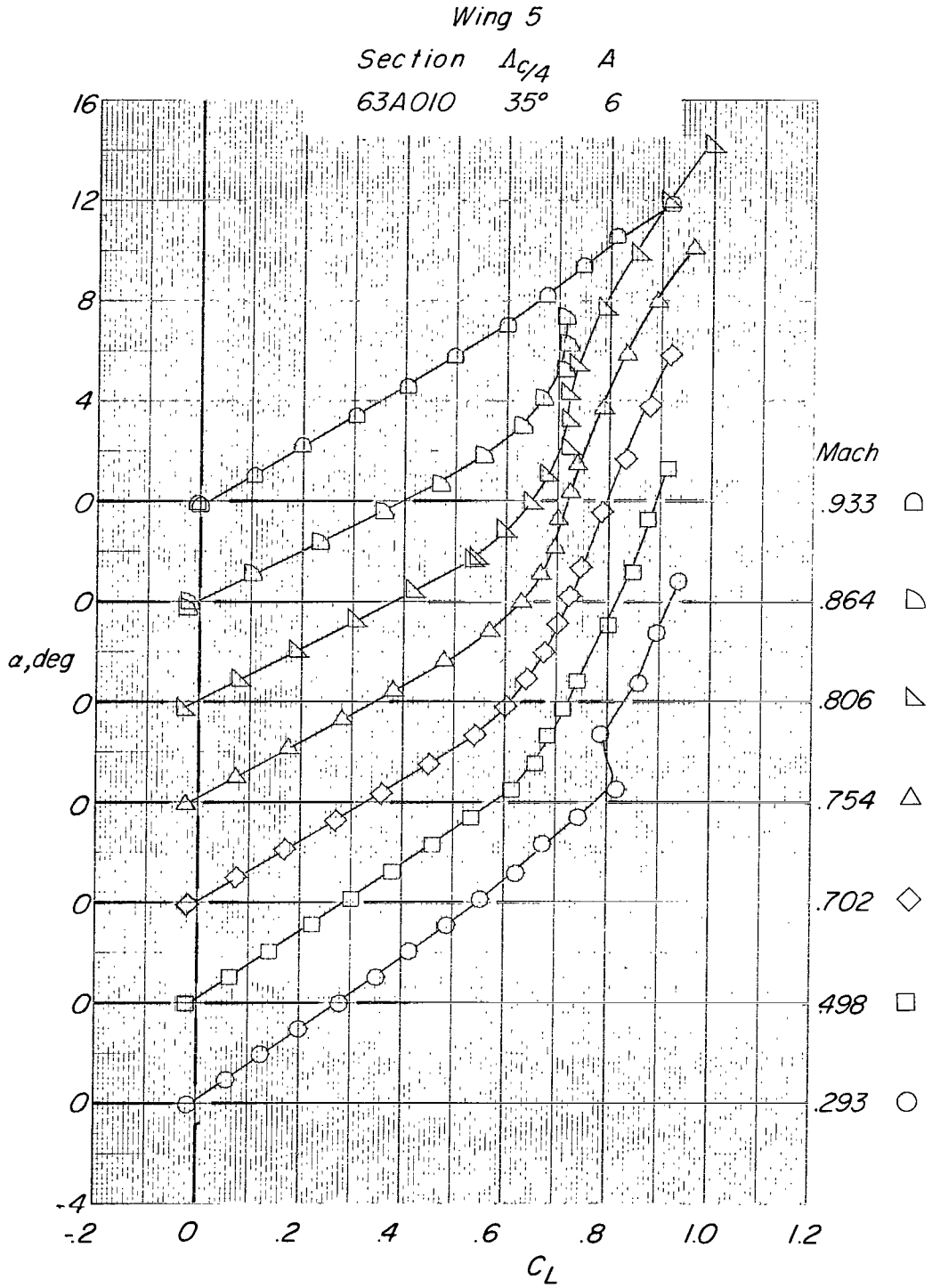
(a) C_N versus α.

Figure 8.- Static longitudinal aerodynamic and buffet characteristics of the wing 5 configuration at Mach numbers from 0.29 to 0.93. Rounded forebody; transition grit on.



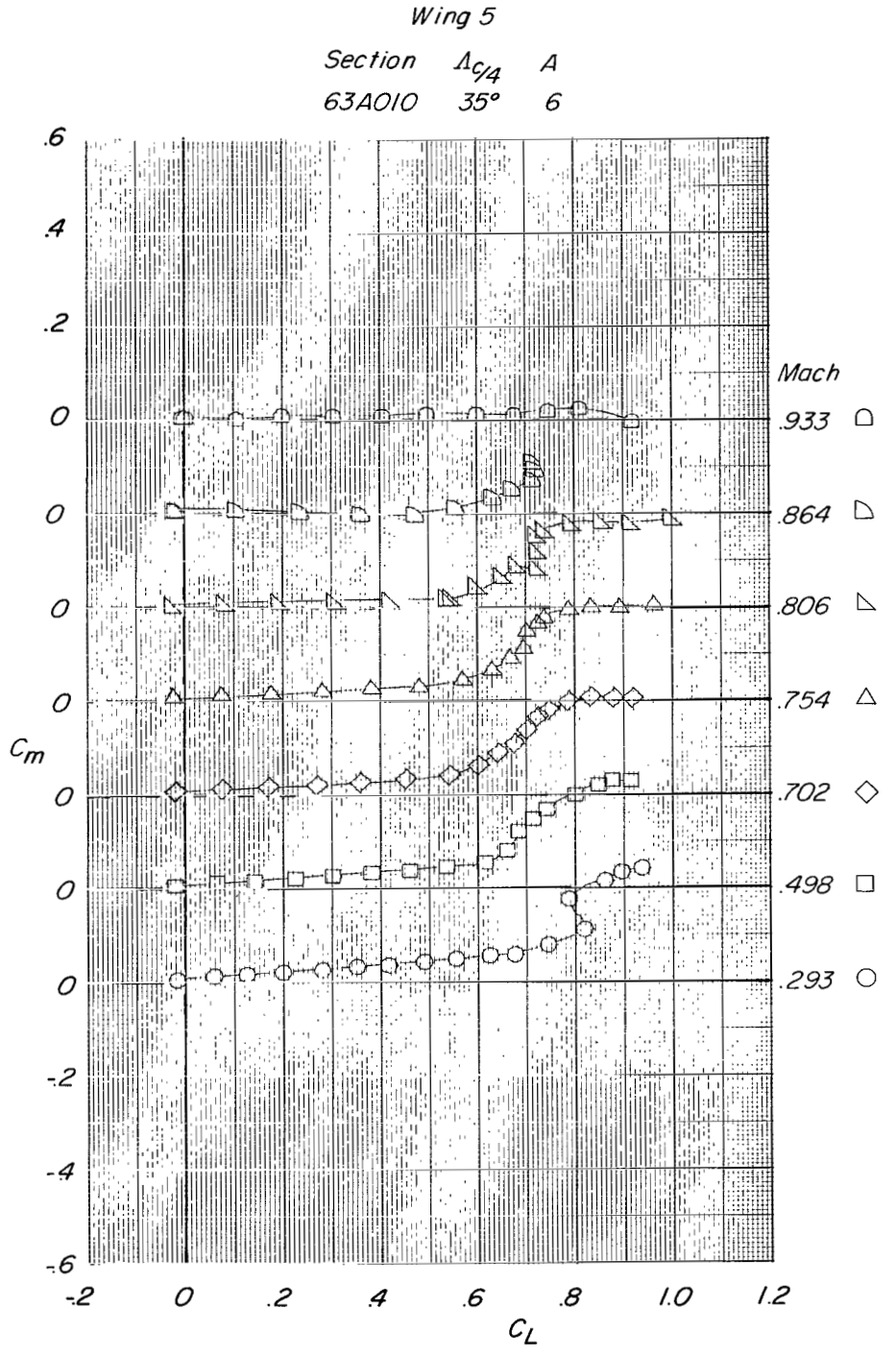
(b) C_A versus α.

Figure 8.- Continued.



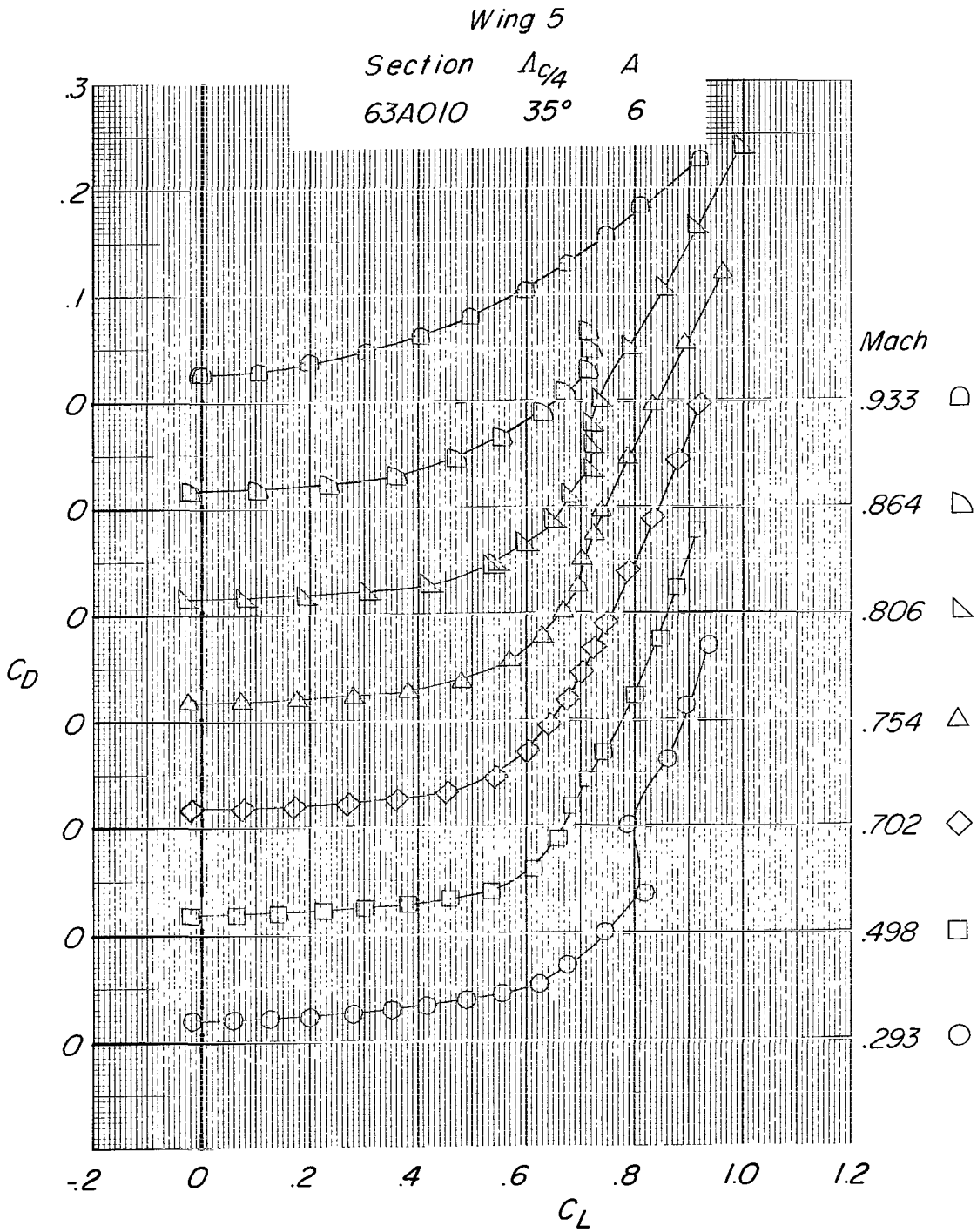
(c) α versus C_L.

Figure 8.- Continued.



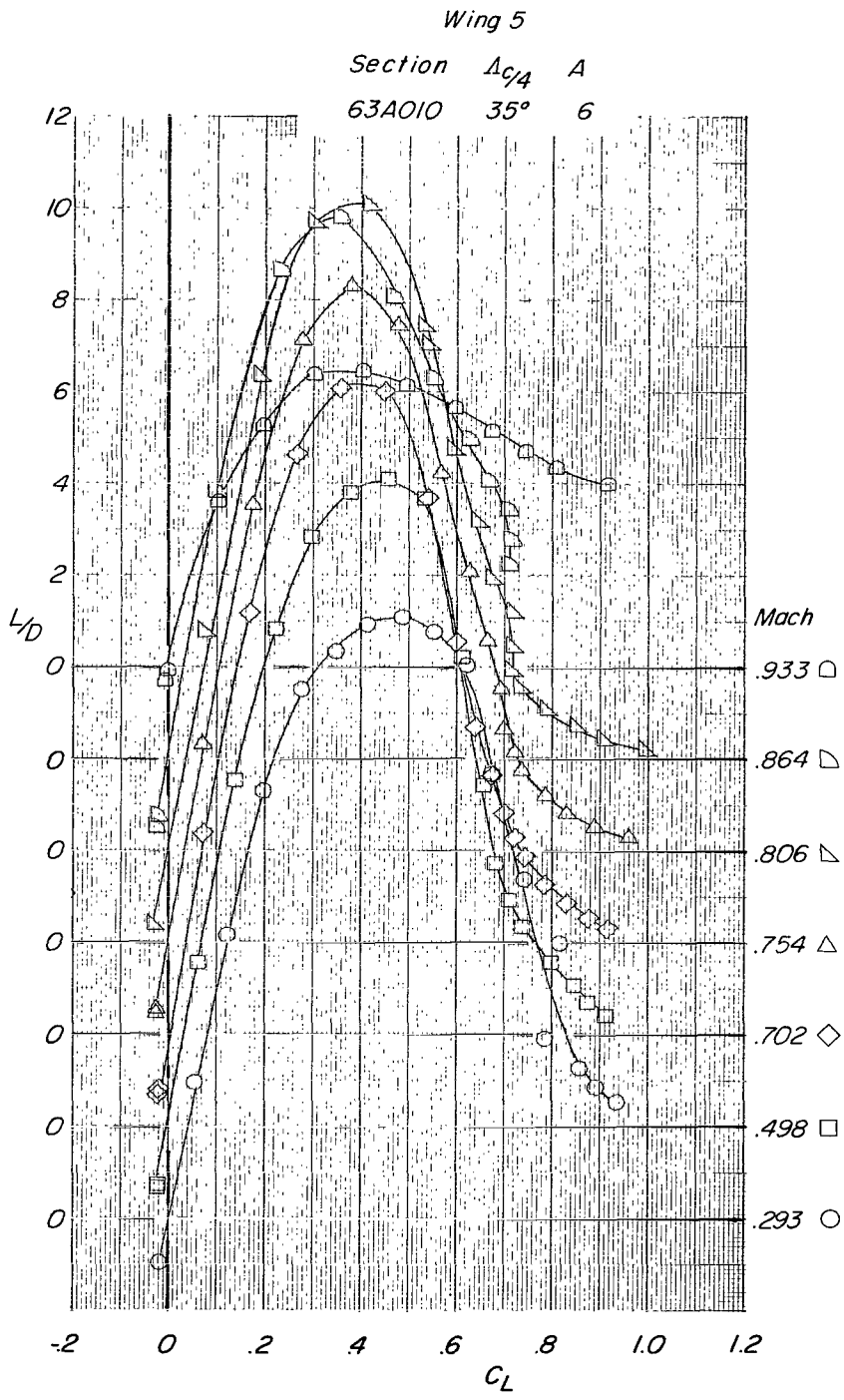
(d) C_m versus C_L.

Figure 8.- Continued.



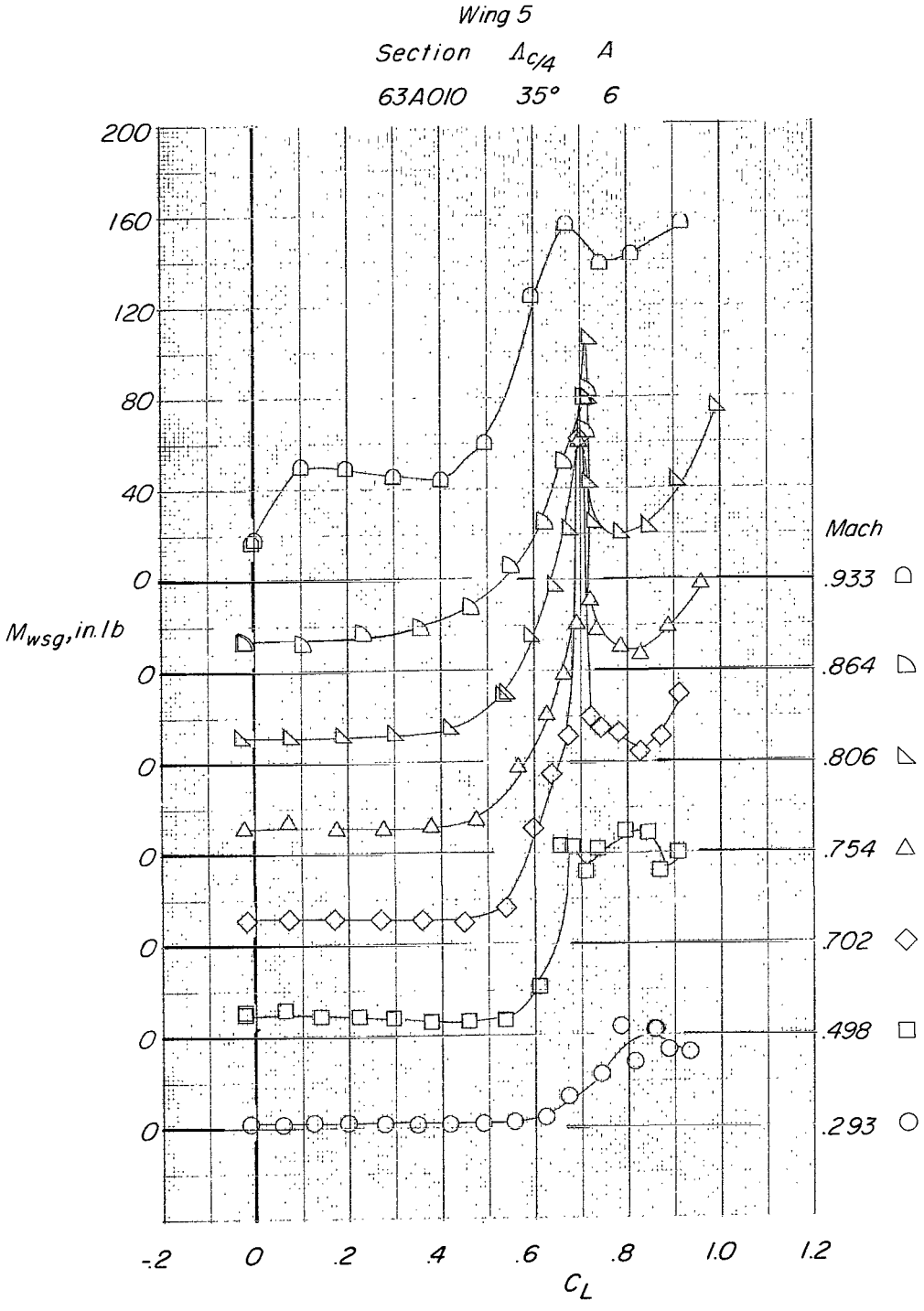
(e) C_D versus C_L.

Figure 8.- Continued.



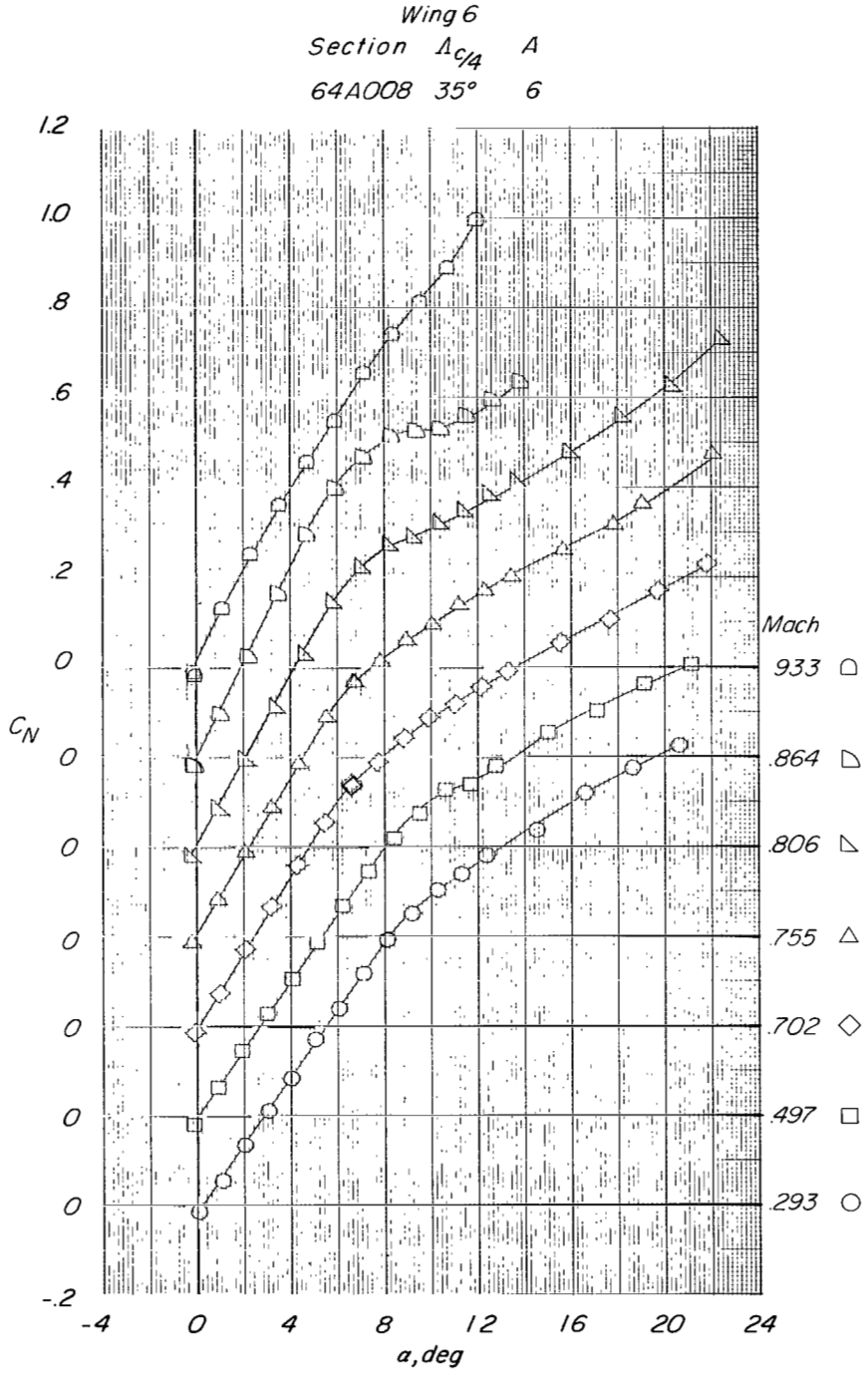
(f) L/D versus C_L.

Figure 8.- Continued.



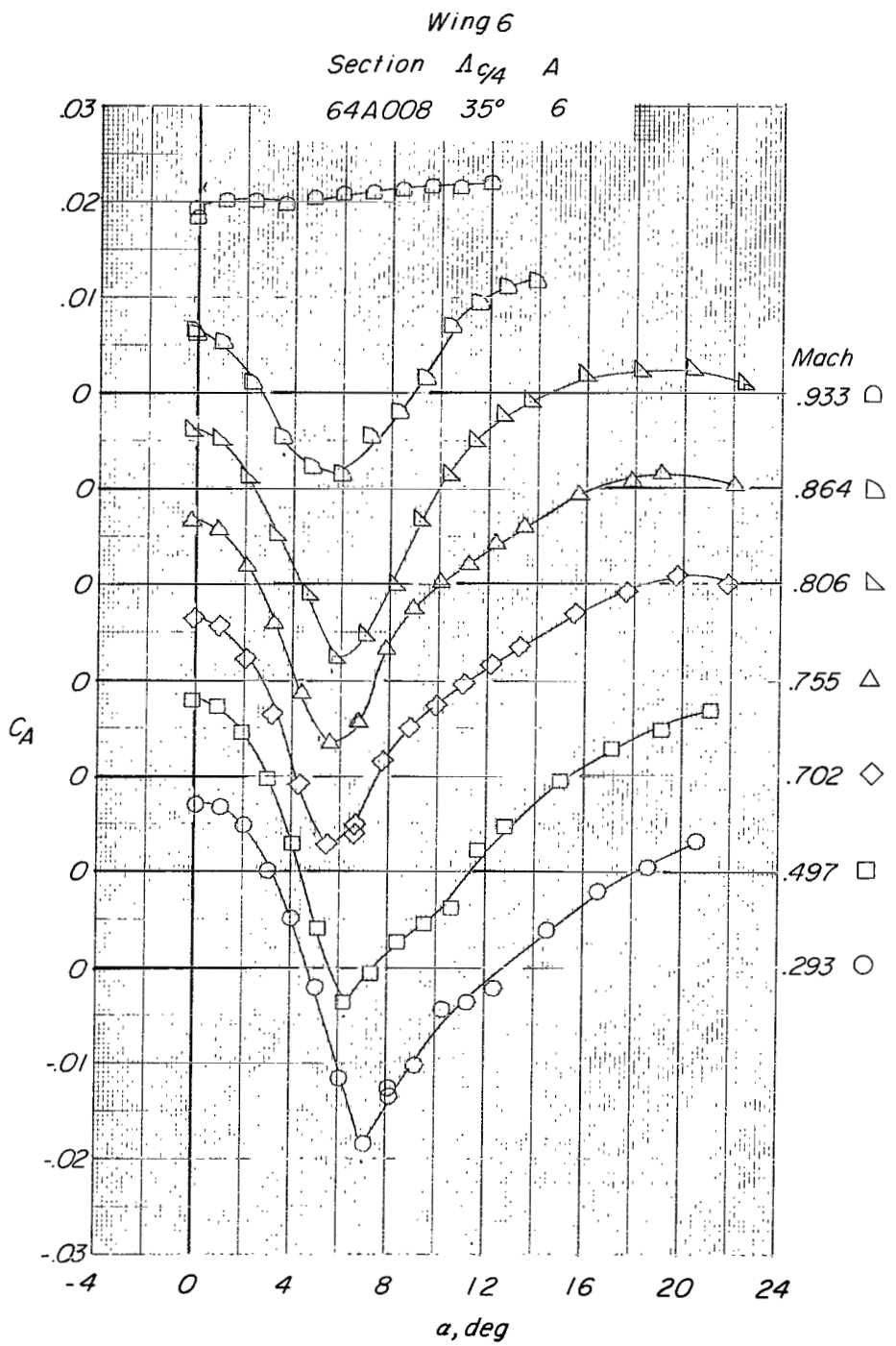
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 8.- Concluded.



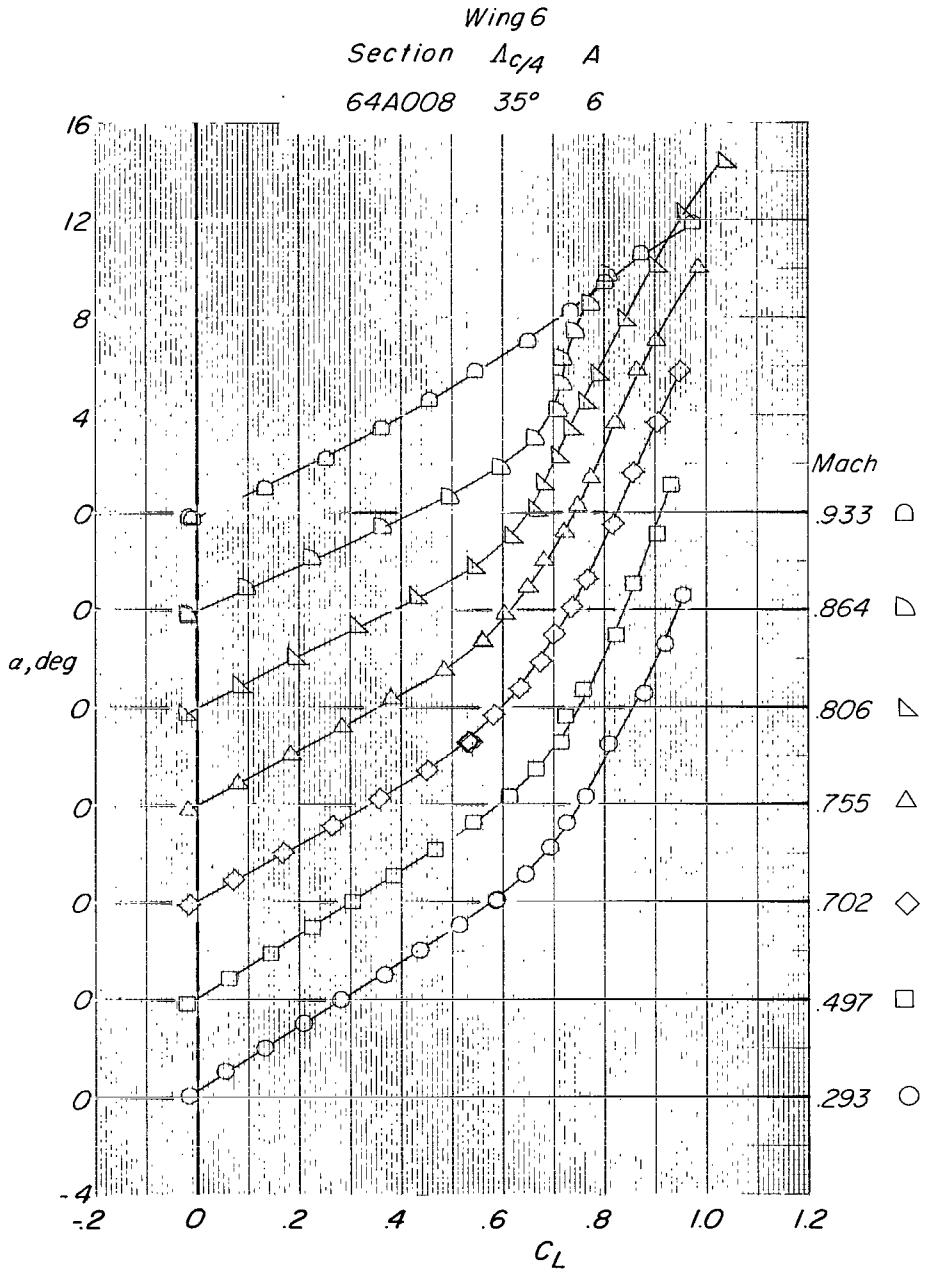
(a) C_N versus α.

Figure 9.- Static longitudinal aerodynamic and buffet characteristics of the wing 6 configuration at Mach numbers from 0.29 to 0.93. Rounded forebody; transition grit on.



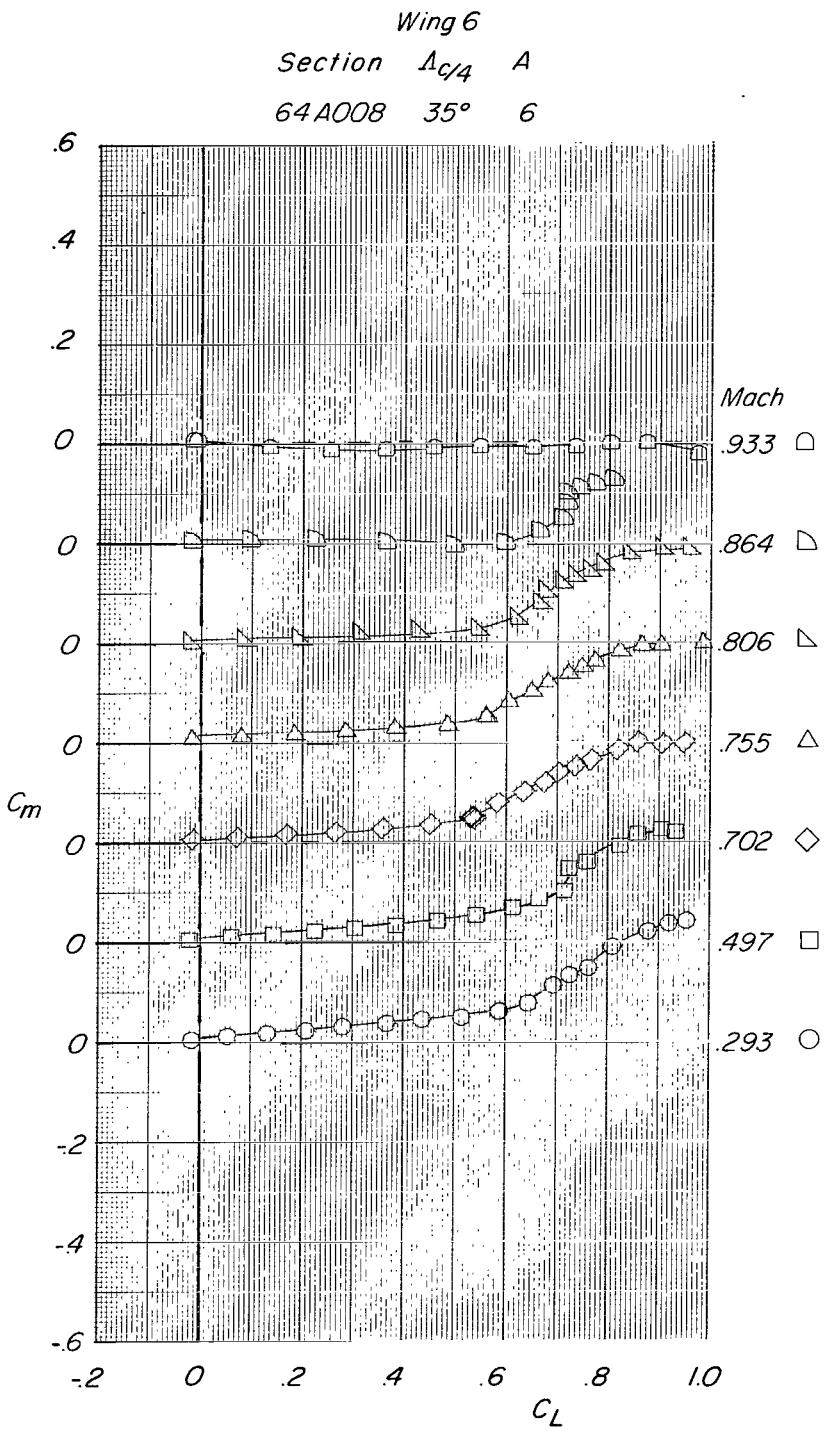
(b) C_A versus α.

Figure 9.- Continued.



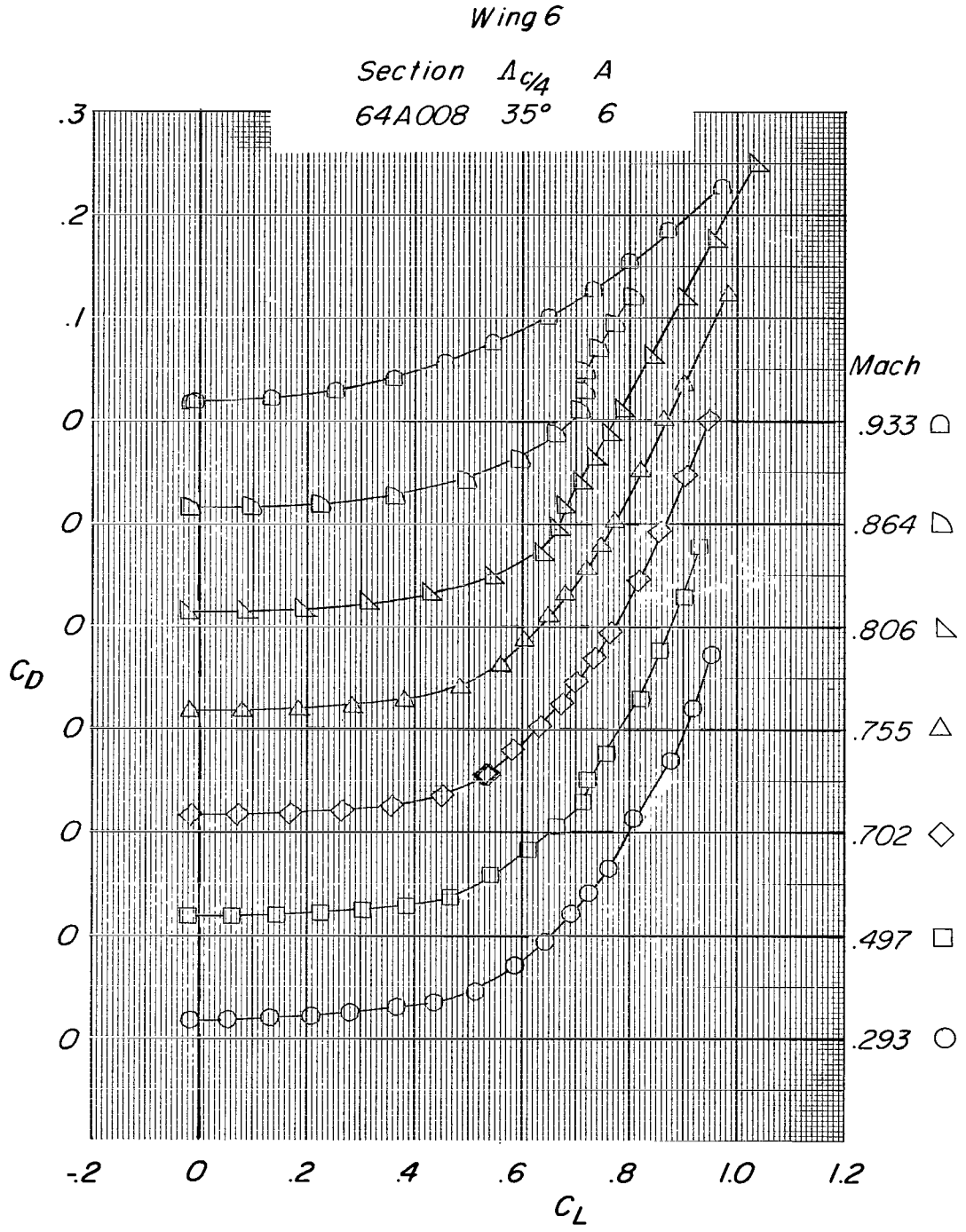
(c) α versus C_L.

Figure 9.- Continued.



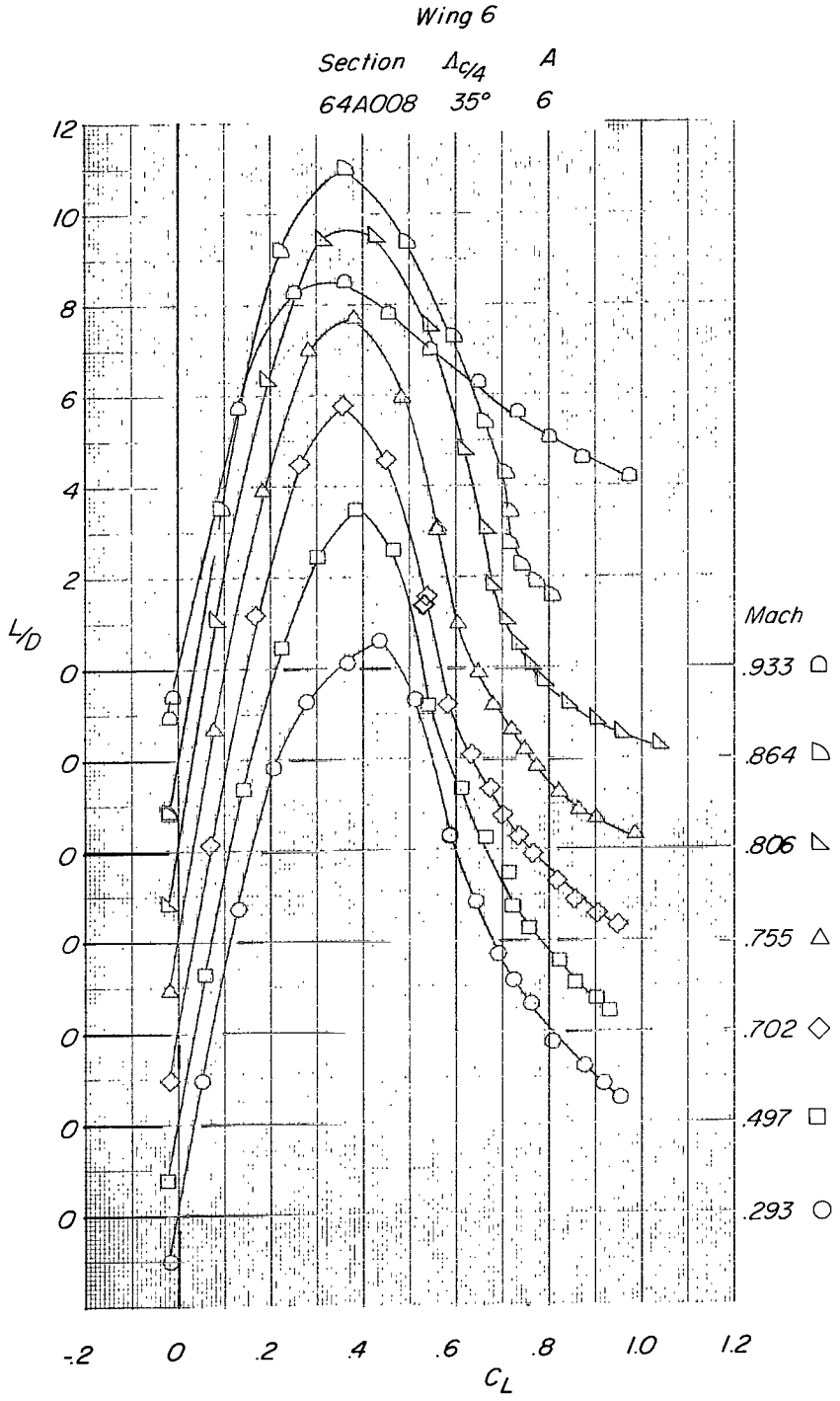
(d) C_m versus C_L.

Figure 9.- Continued.



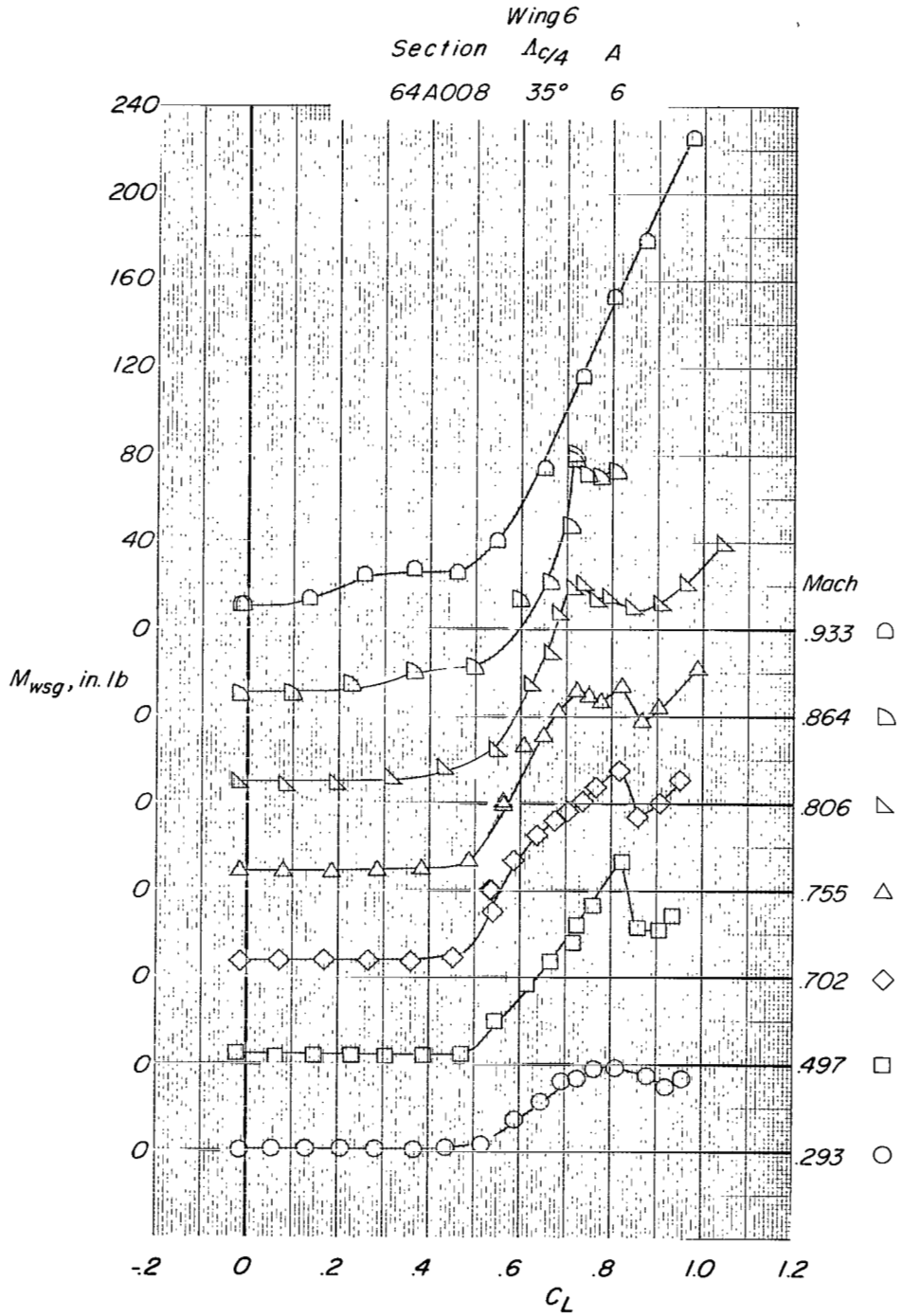
(e) C_D versus C_L.

Figure 9.- Continued.



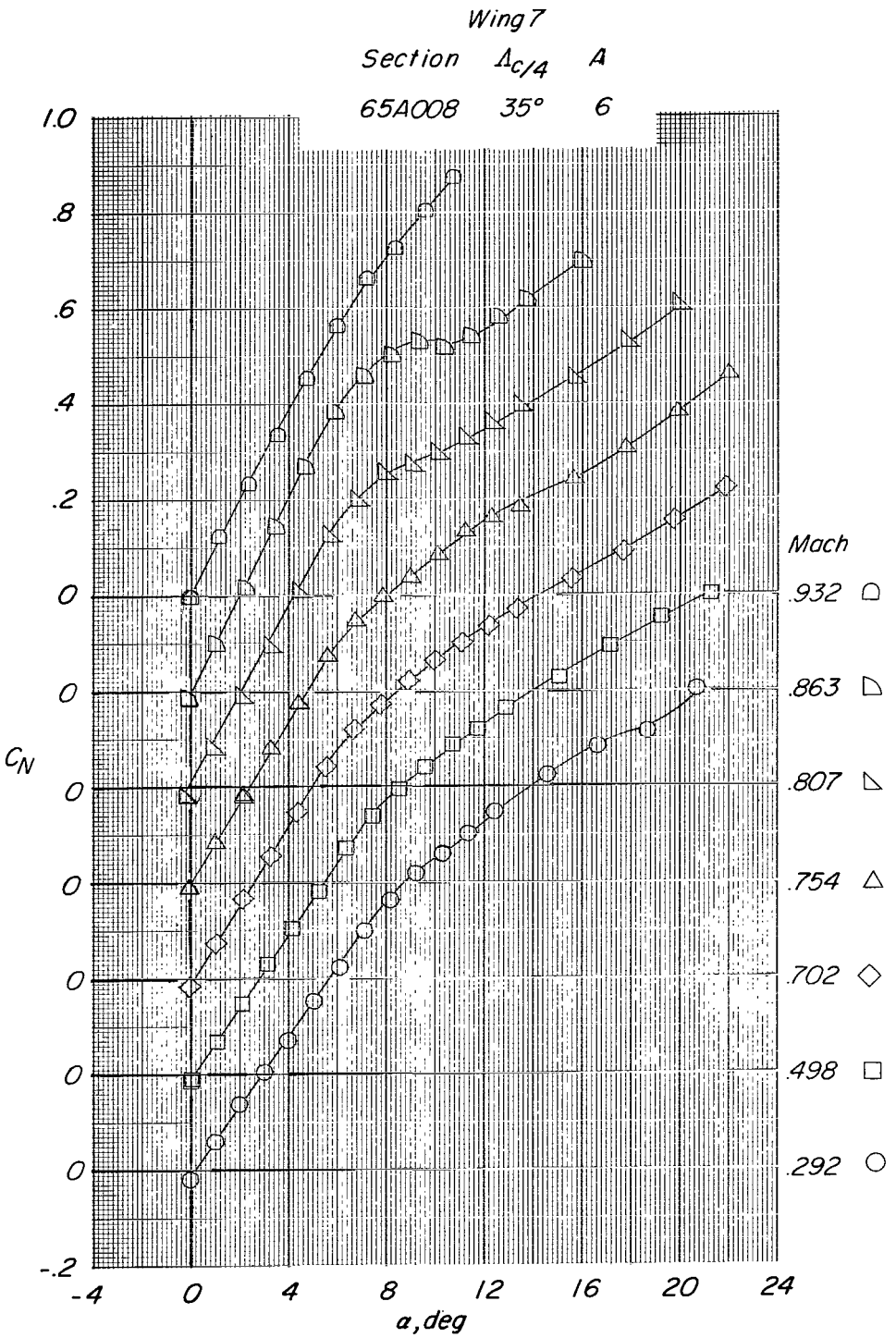
(f) L/D versus C_L.

Figure 9.- Continued.



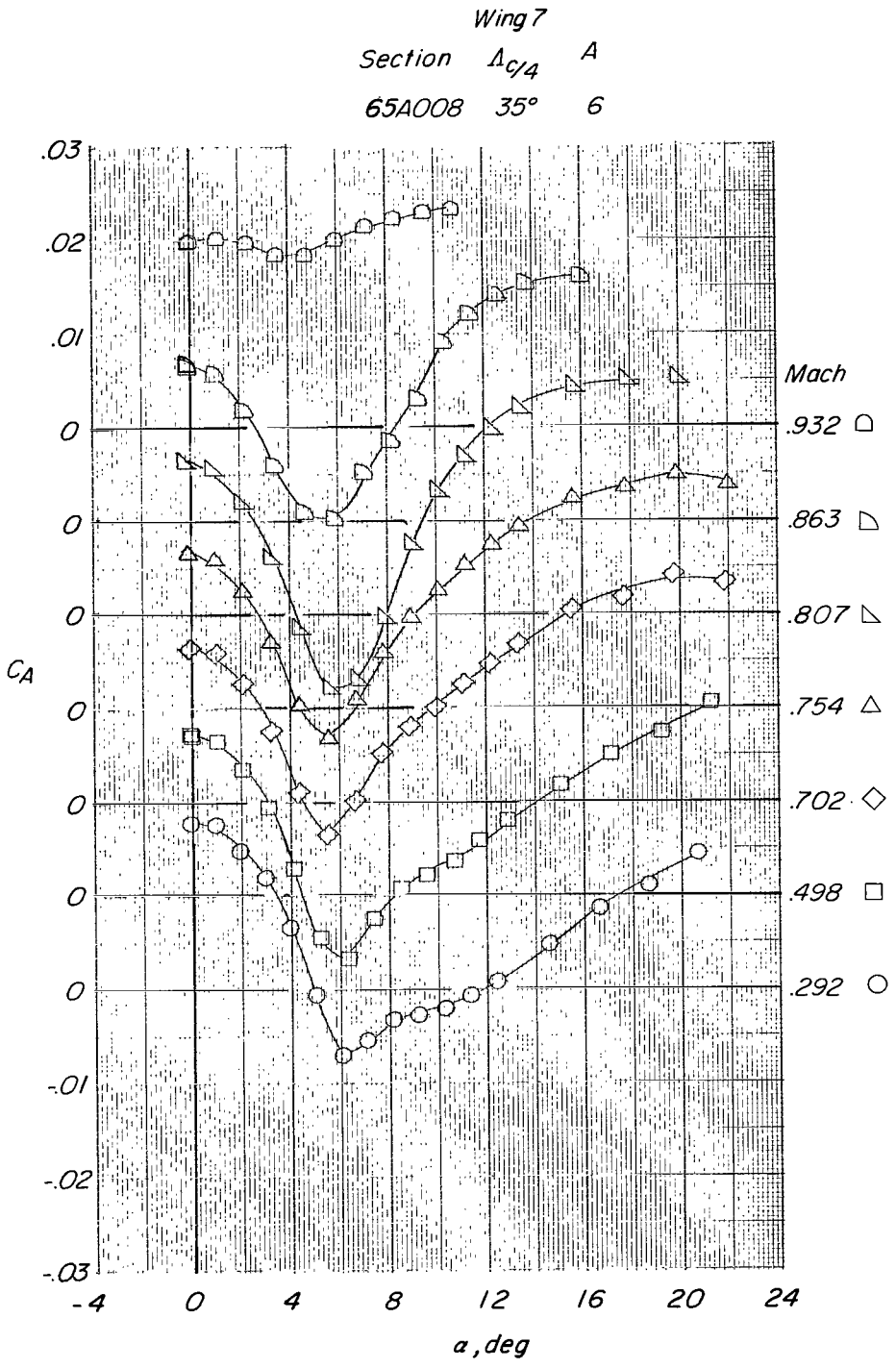
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 9.- Concluded.



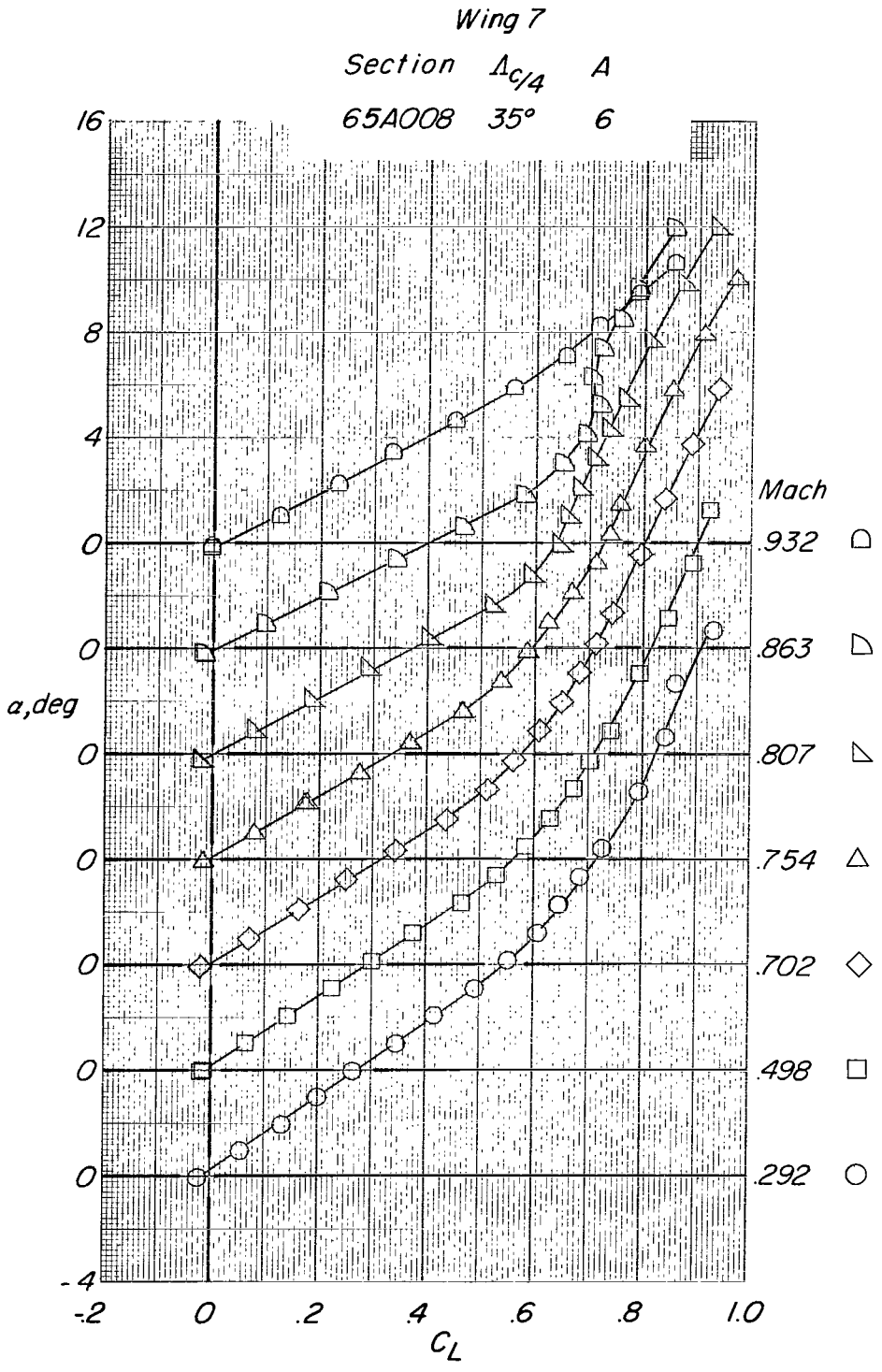
(a) C_L versus α.

Figure 10.- Static longitudinal aerodynamic and buffet characteristics of the wing 7 configuration at Mach numbers from 0.29 to 0.93. Rounded forebody; transition grit on.



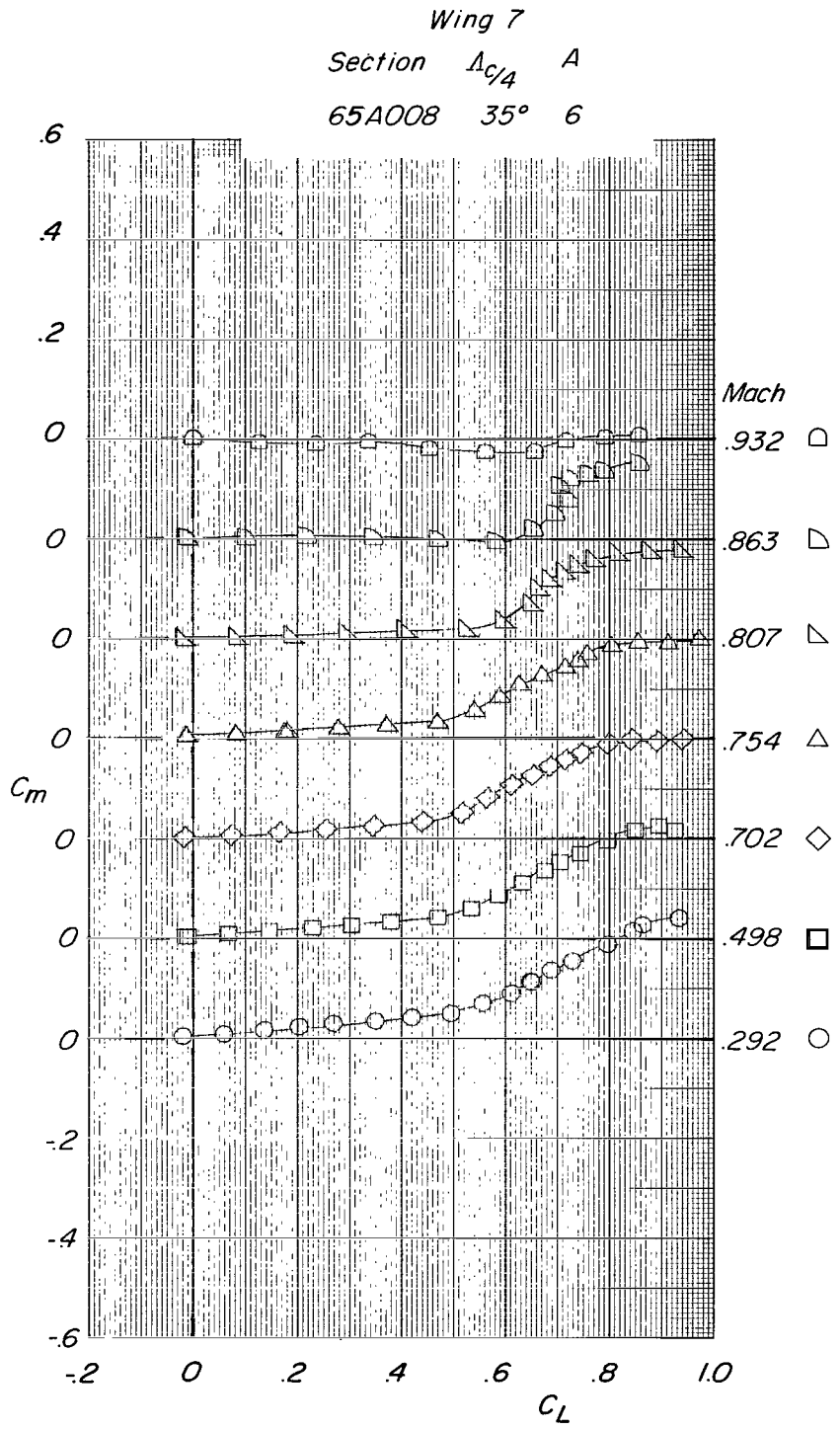
(b) C_A versus α.

Figure 10.- Continued.



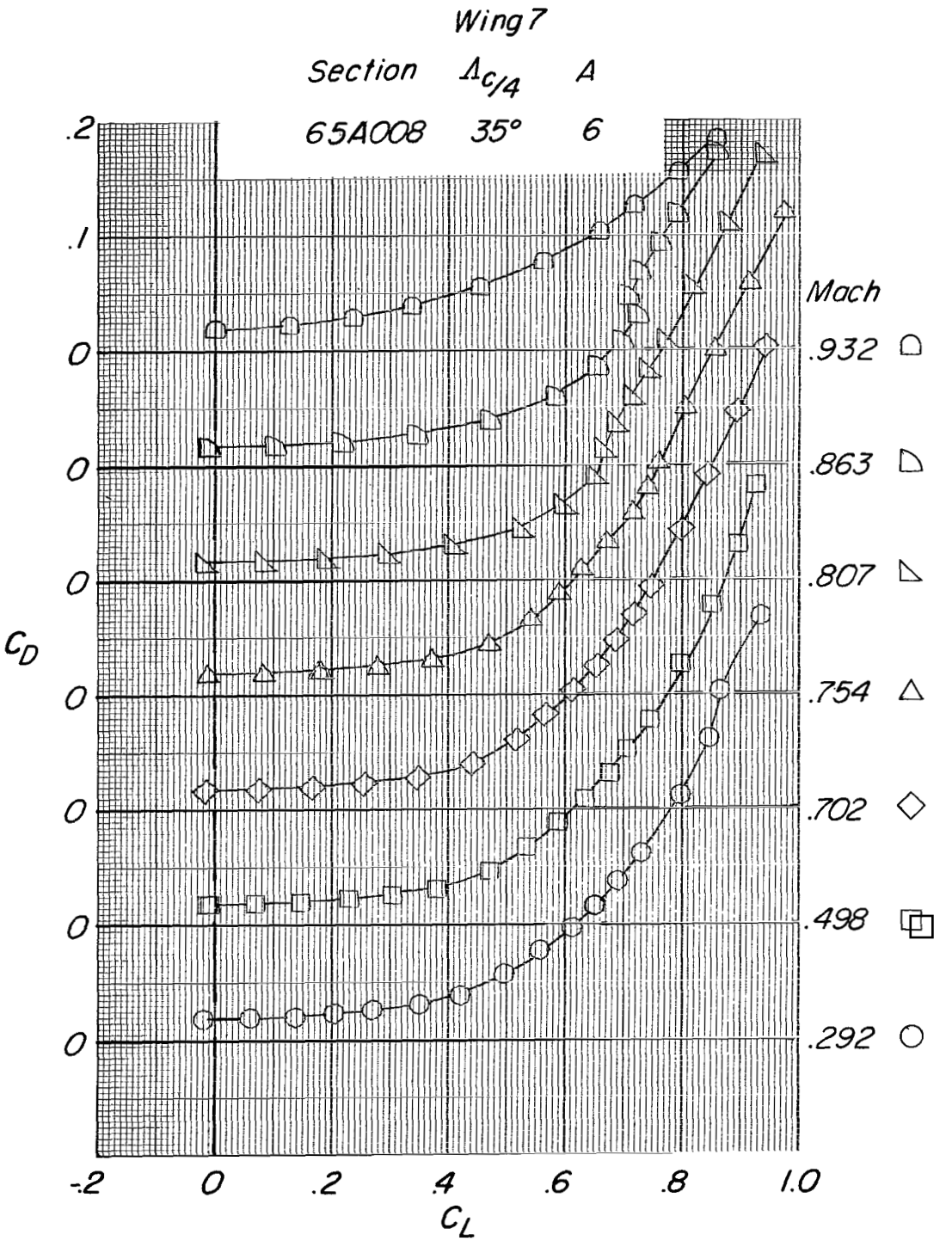
(c) α versus C_L.

Figure 10.- Continued.



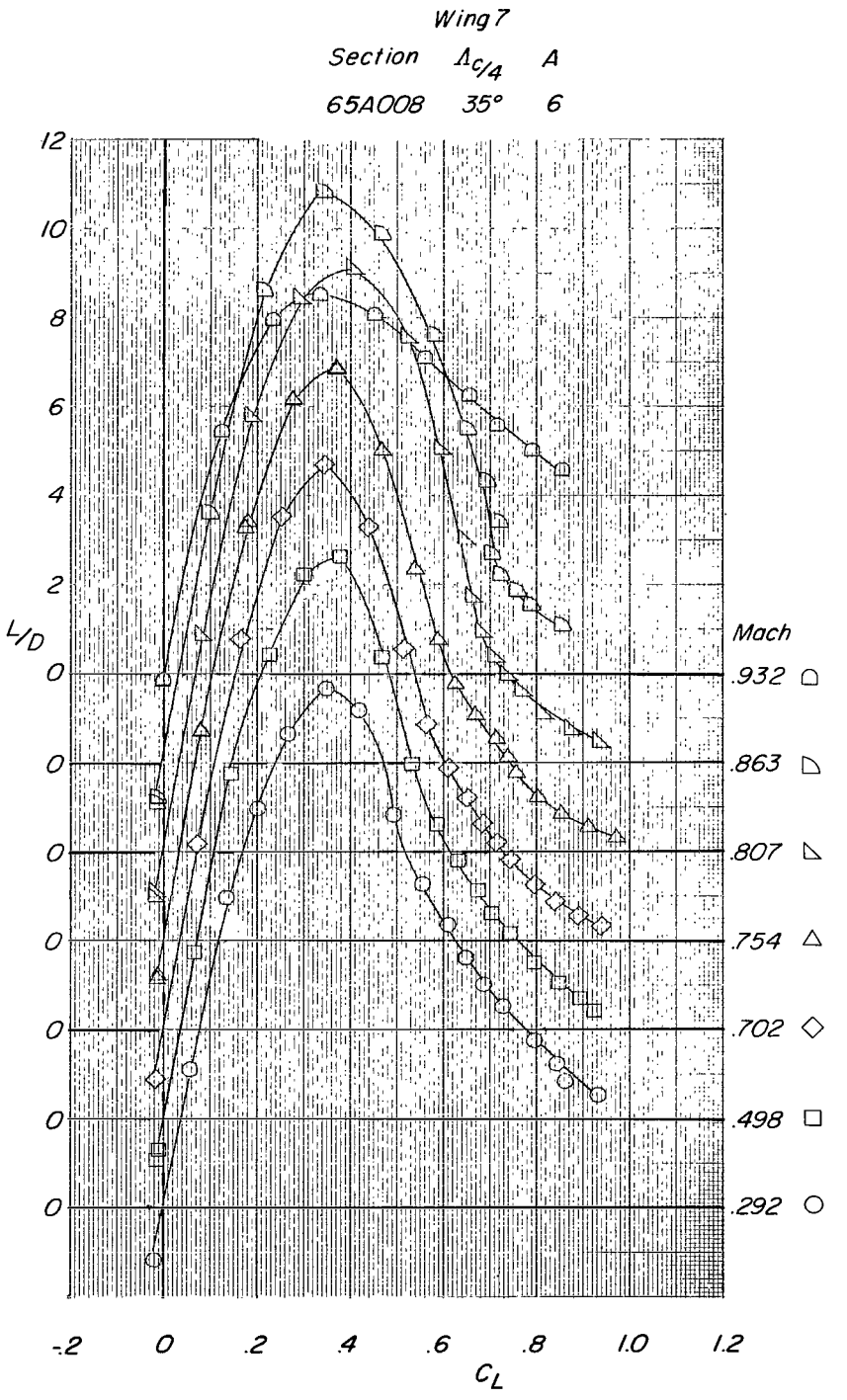
(d) C_m versus C_L.

Figure 10.- Continued.



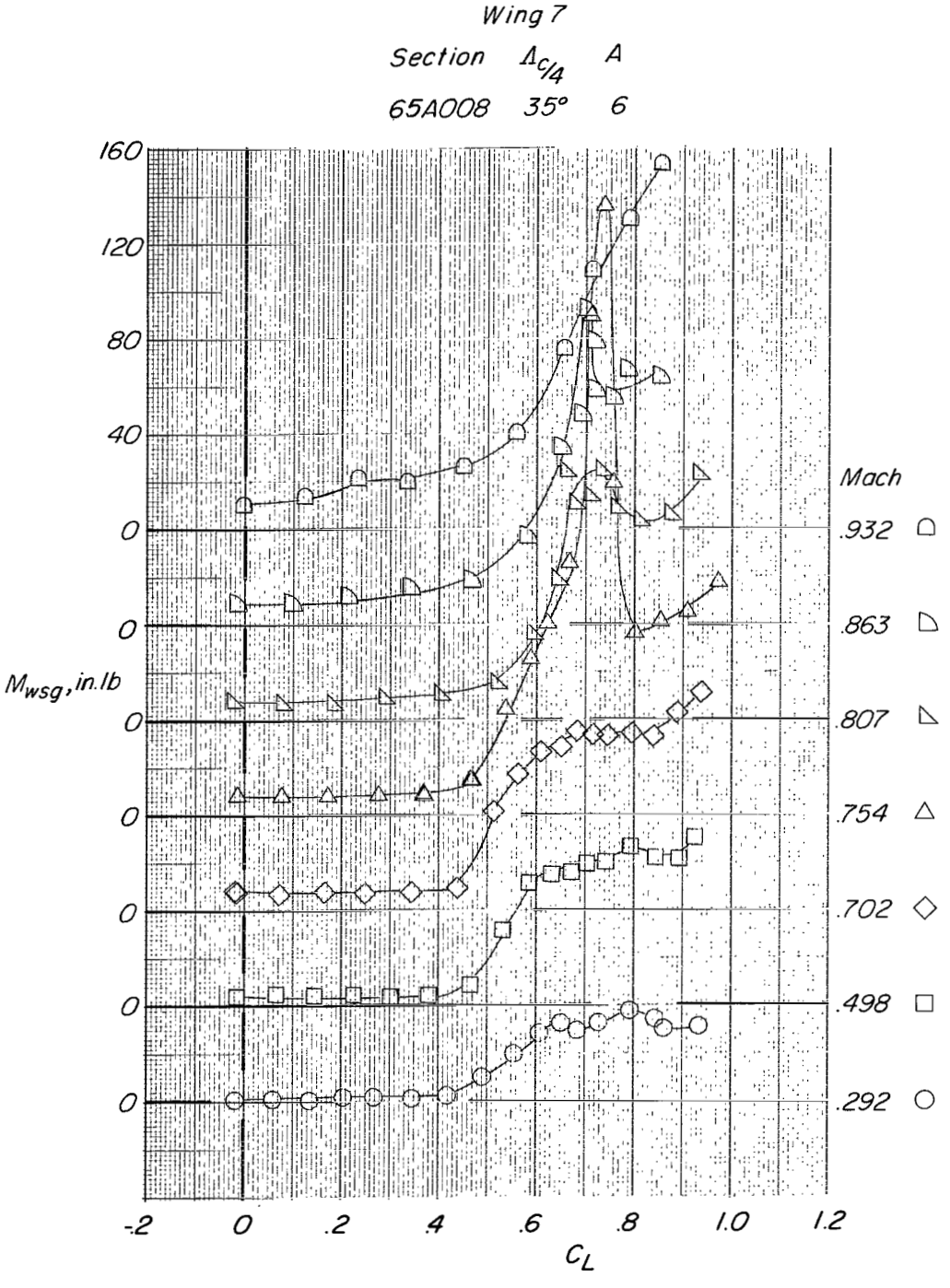
(e) C_D versus C_L.

Figure 10.- Continued.



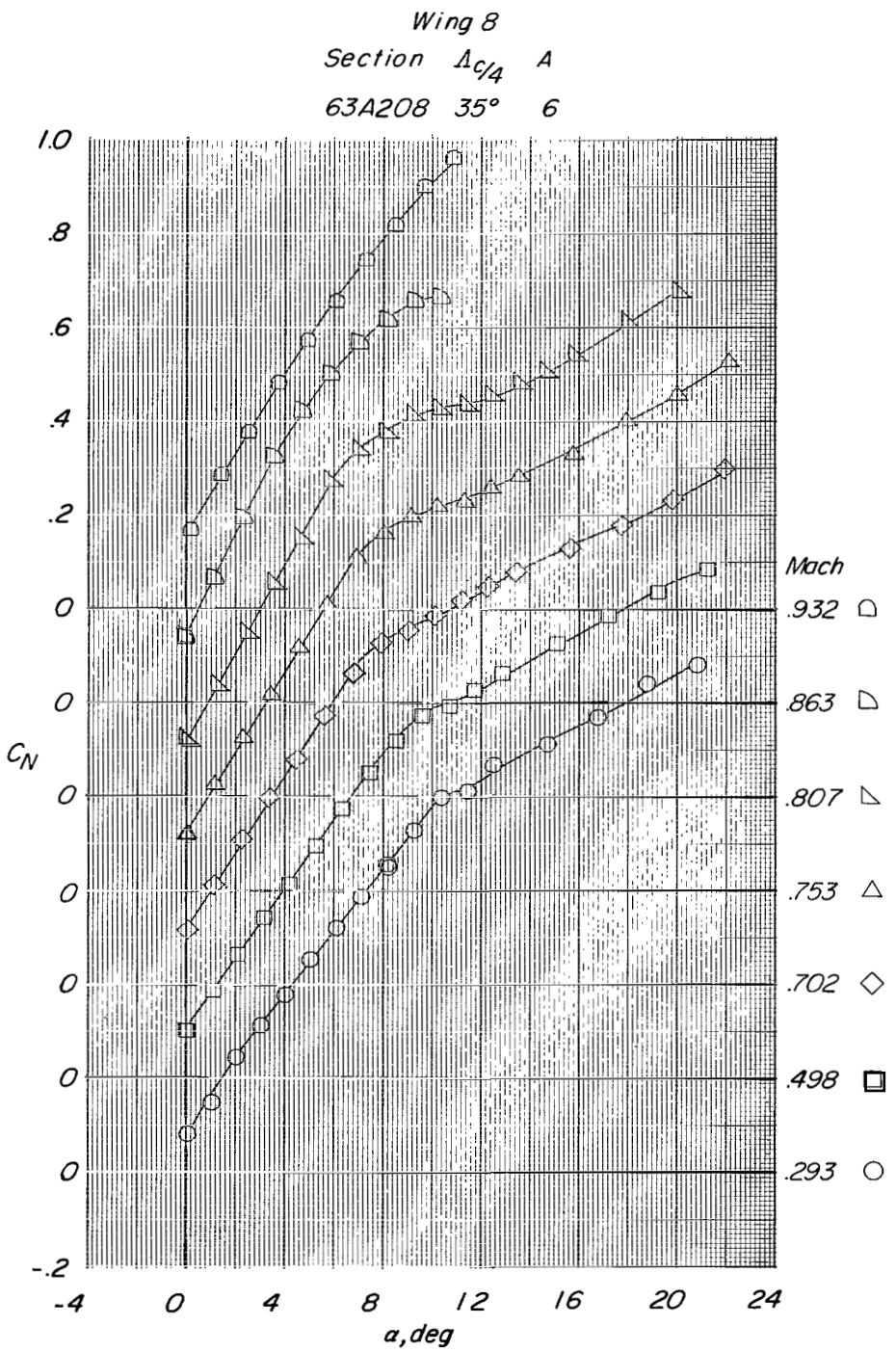
(f) L/D versus C_L.

Figure 10.- Continued.



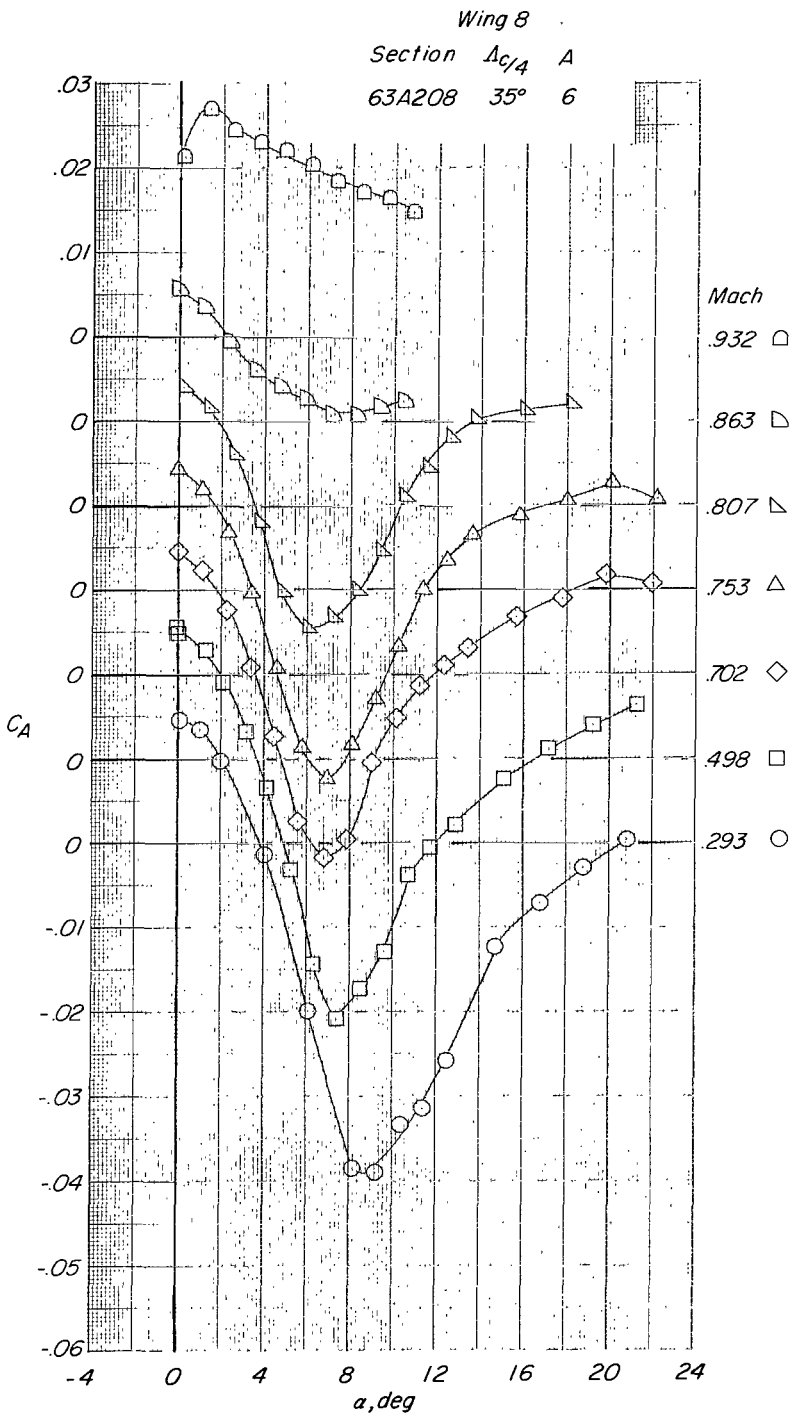
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 10.- Concluded.



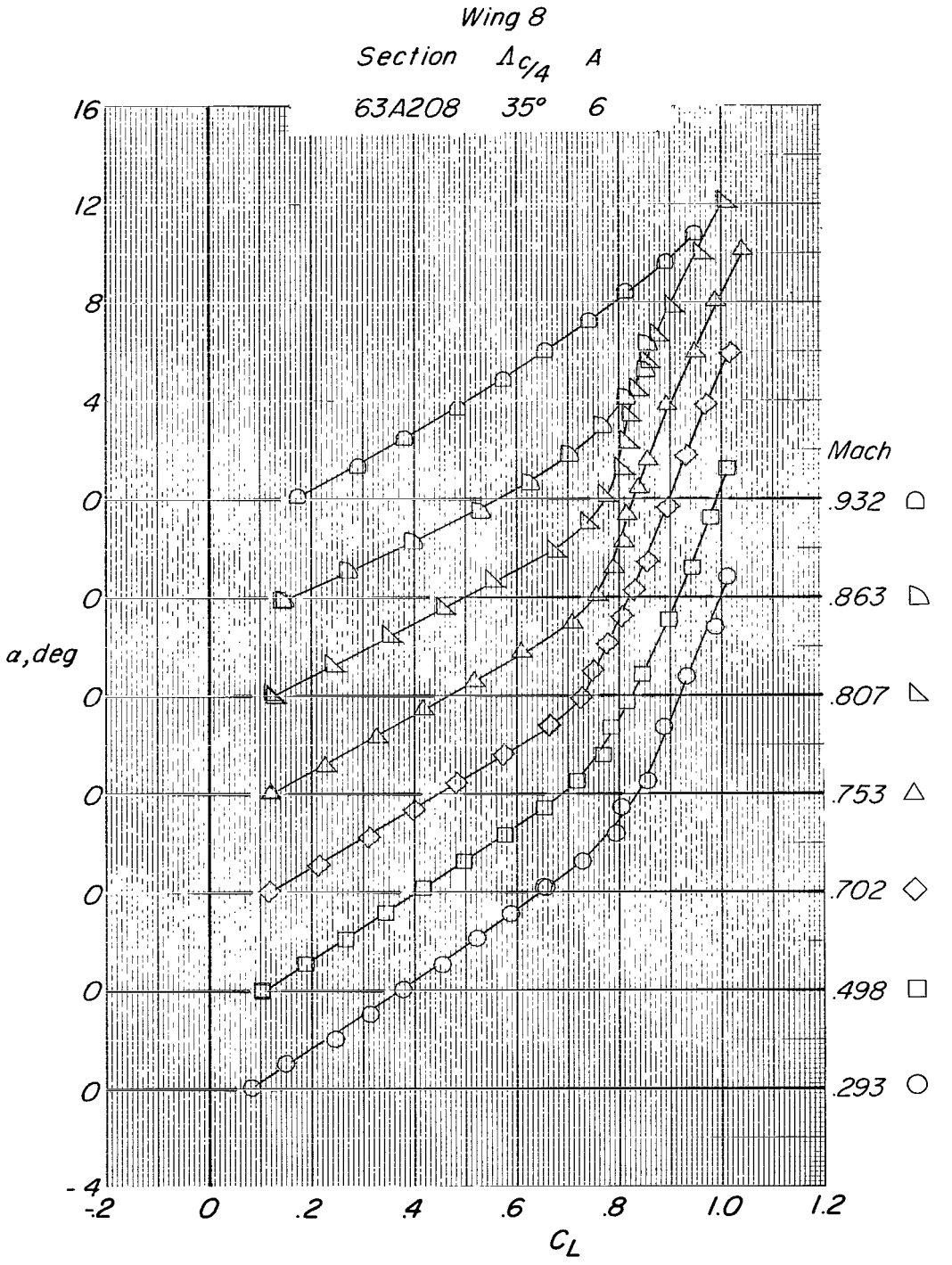
(a) C_N versus α.

Figure 11.- Static longitudinal aerodynamic and buffet characteristics of the wing 8 configuration at Mach numbers from 0.29 to 0.93. Rounded forebody; transition grit off.



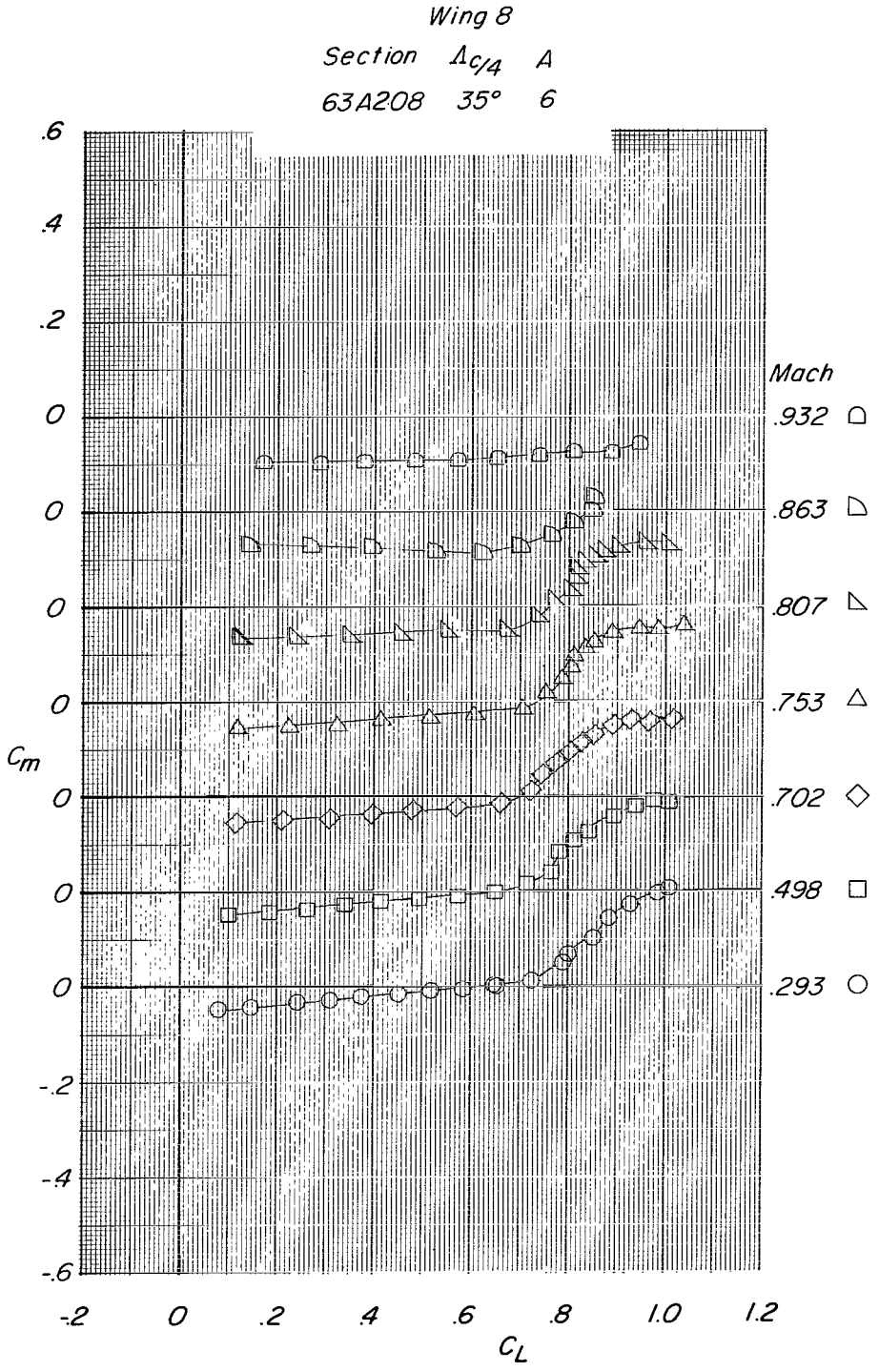
(b) C_A versus α.

Figure 11.- Continued.



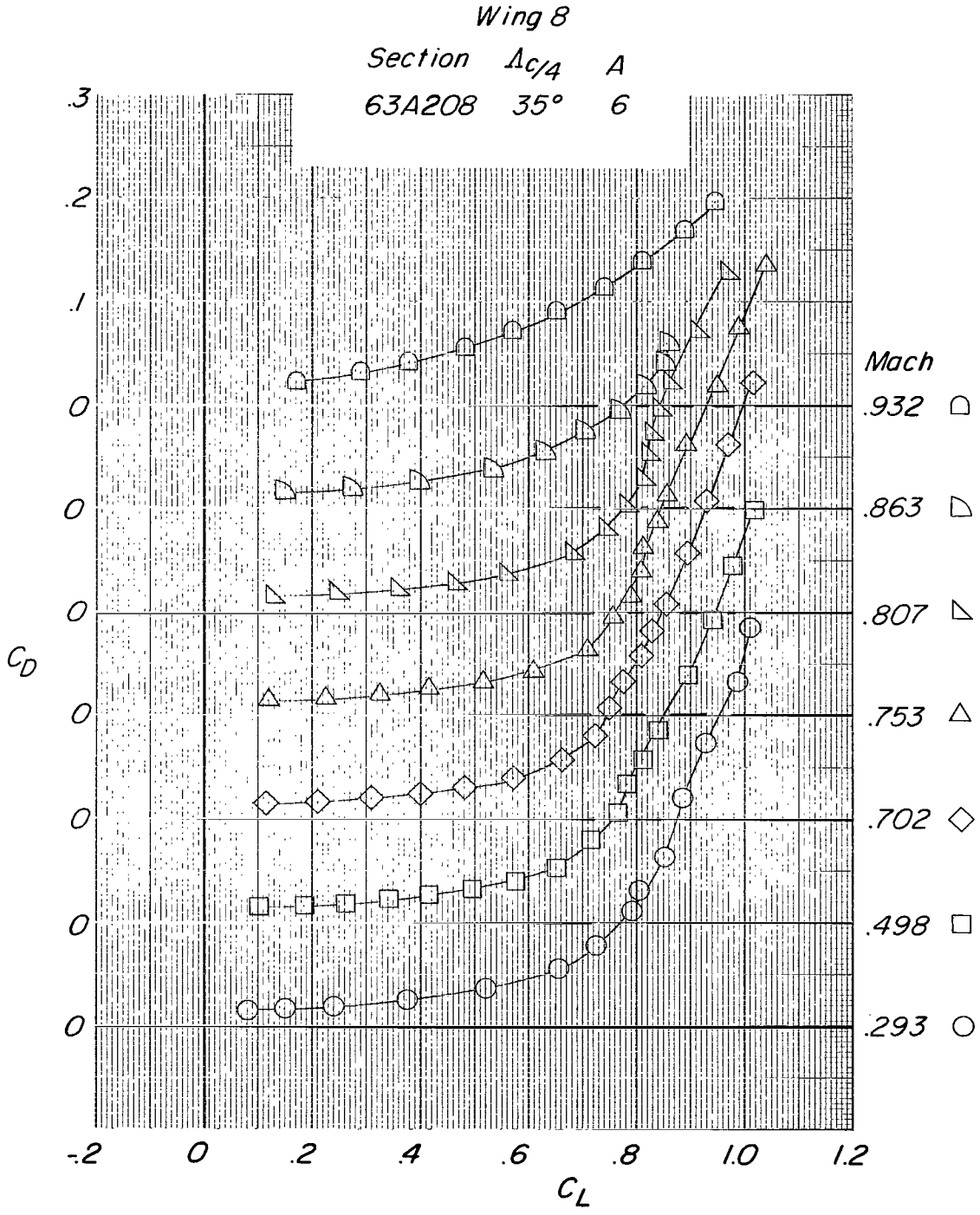
(c) α versus C_L.

Figure 11.- Continued.



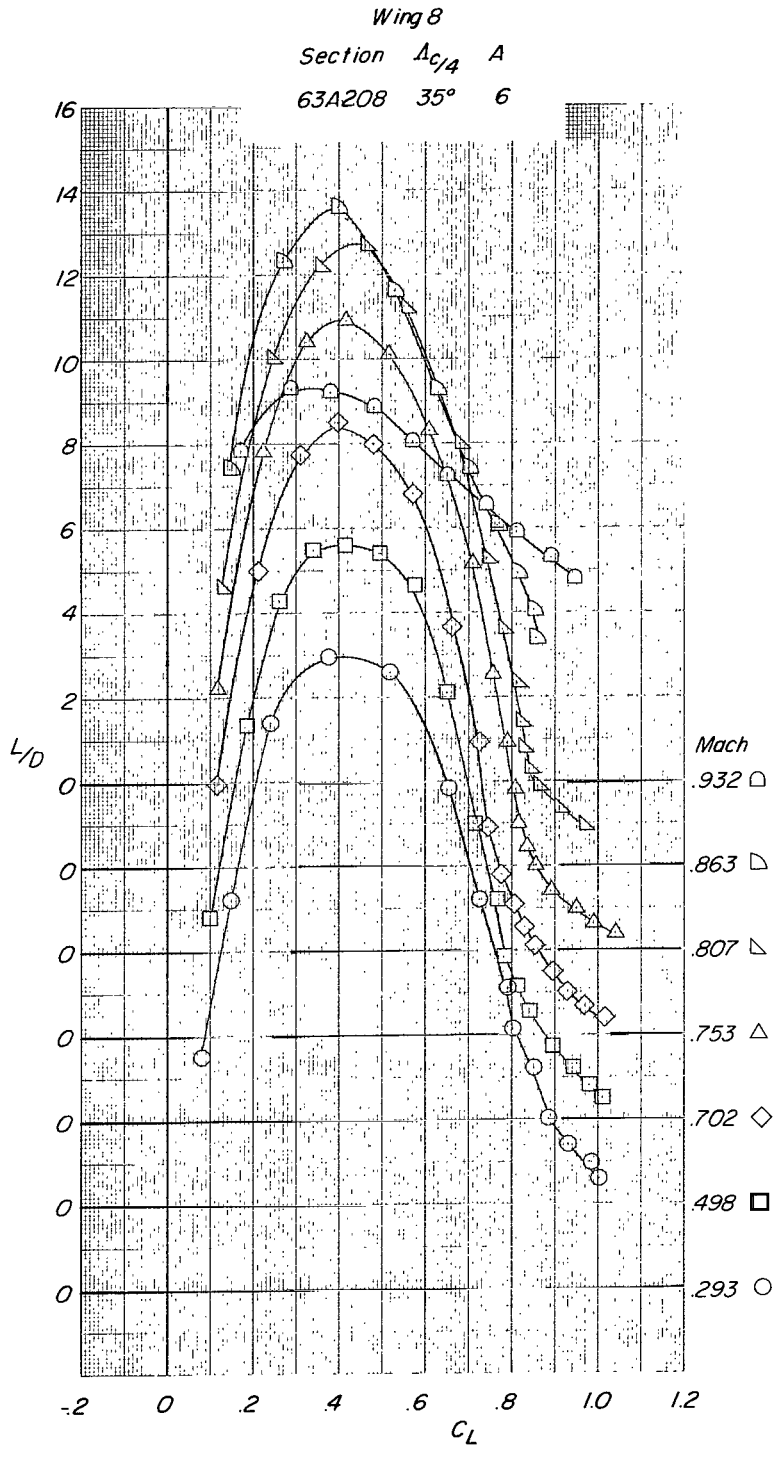
(d) C_m versus C_L.

Figure 11.- Continued.



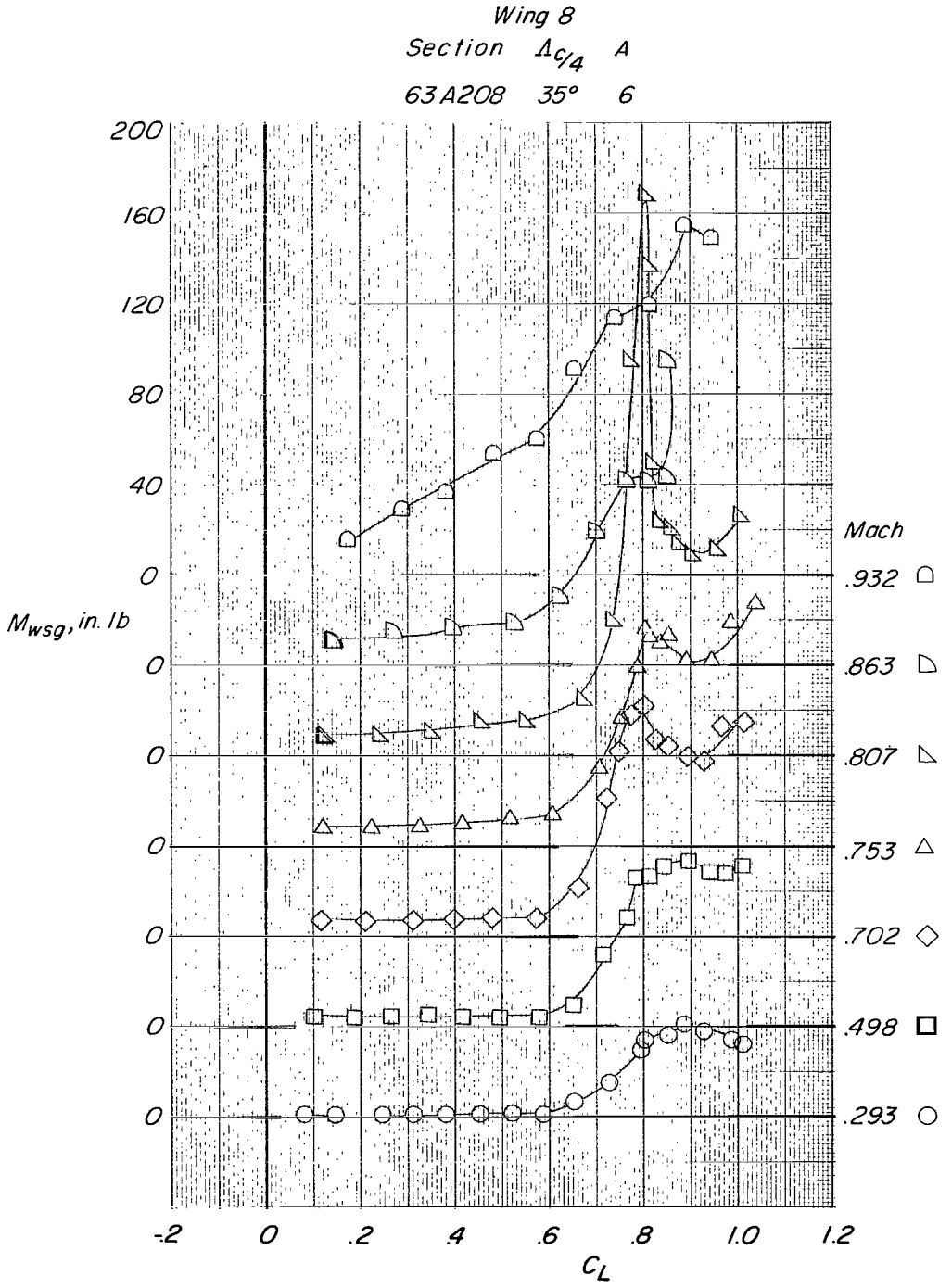
(e) C_D versus C_L.

Figure 11.- Continued.



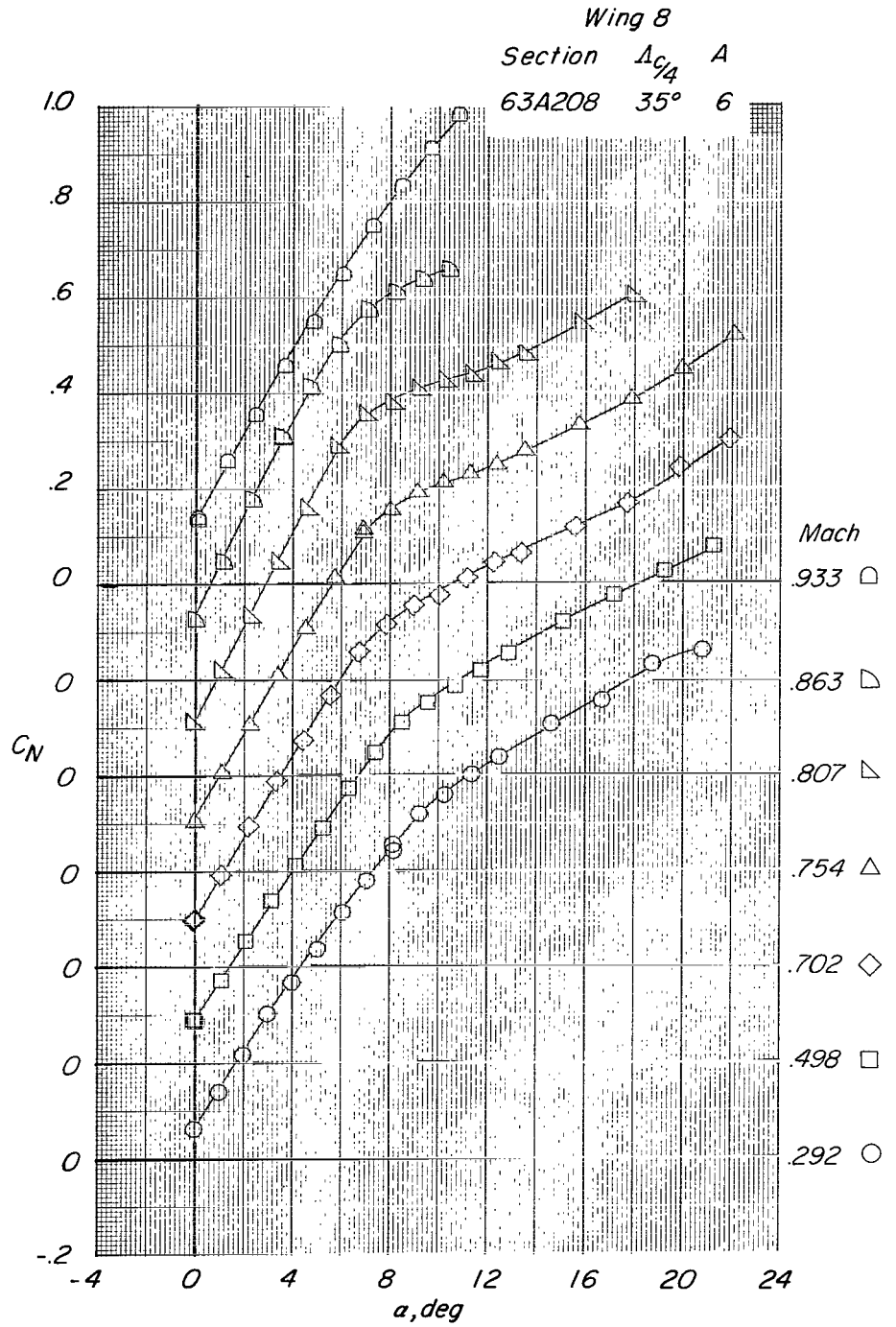
(f) L/D versus C_L.

Figure 11.- Continued.



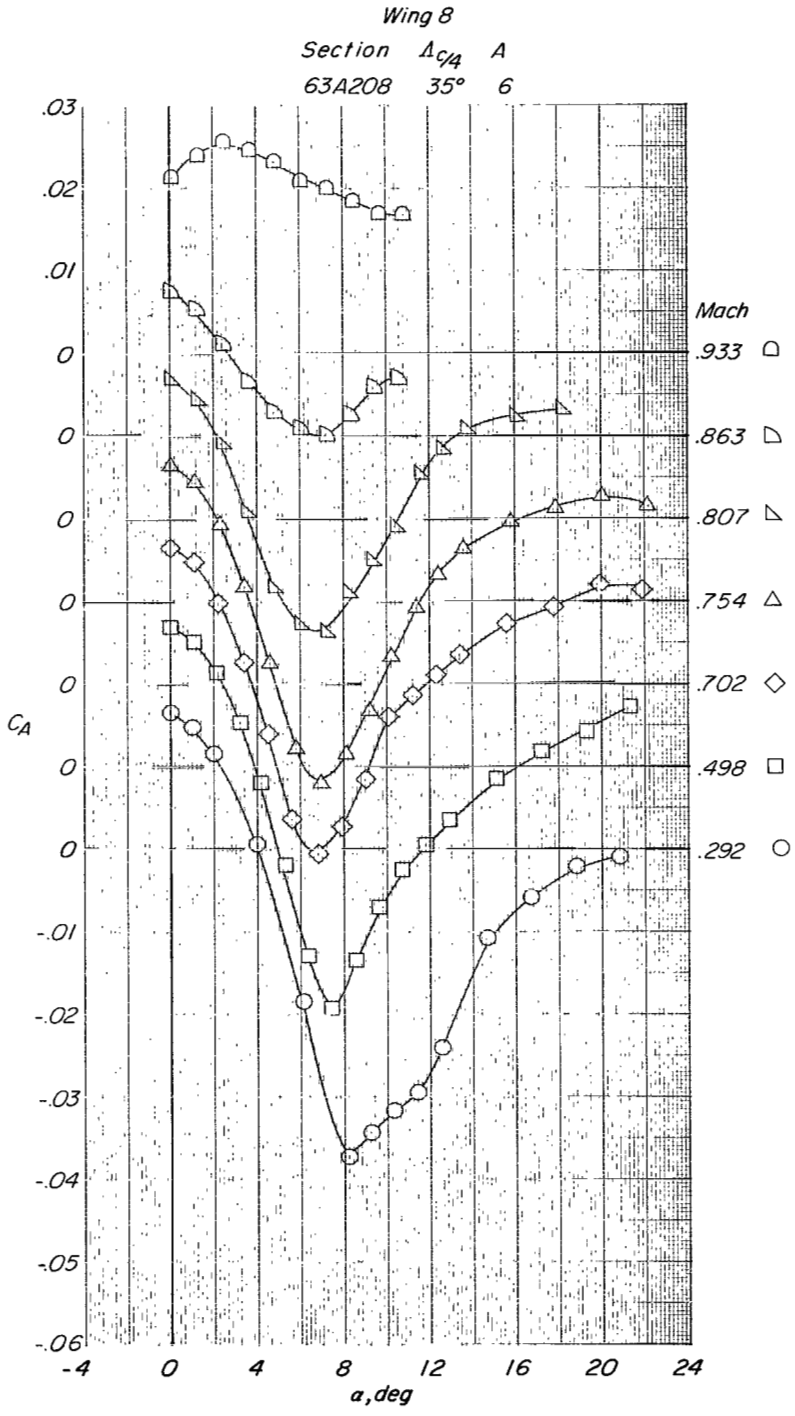
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 11.- Concluded.



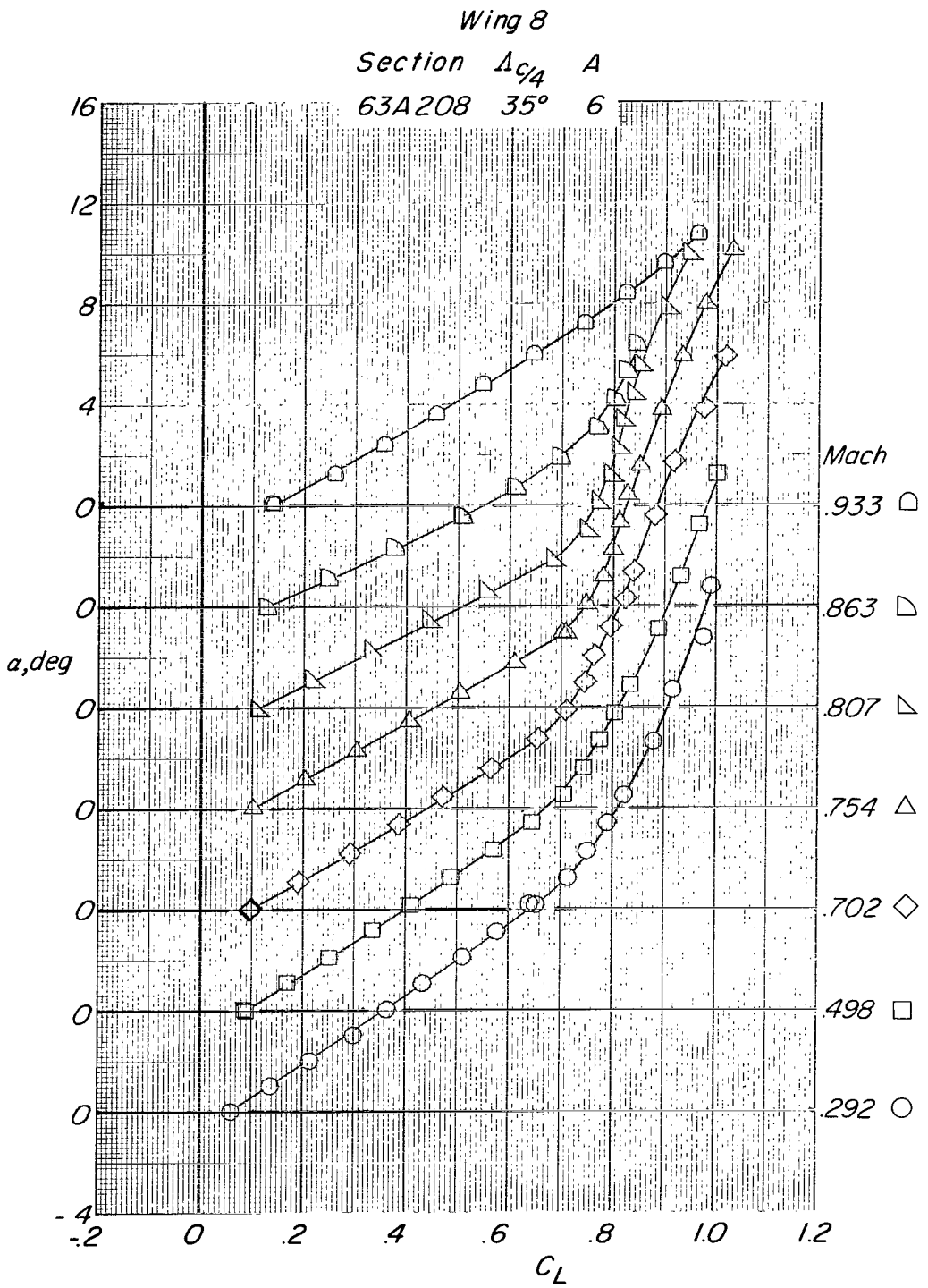
(a) C_N versus α.

Figure 12.- Static longitudinal aerodynamic and buffet characteristics of the wing 8 configuration at Mach numbers from 0.29 to 0.93. Rounded forebody; transition grit on.



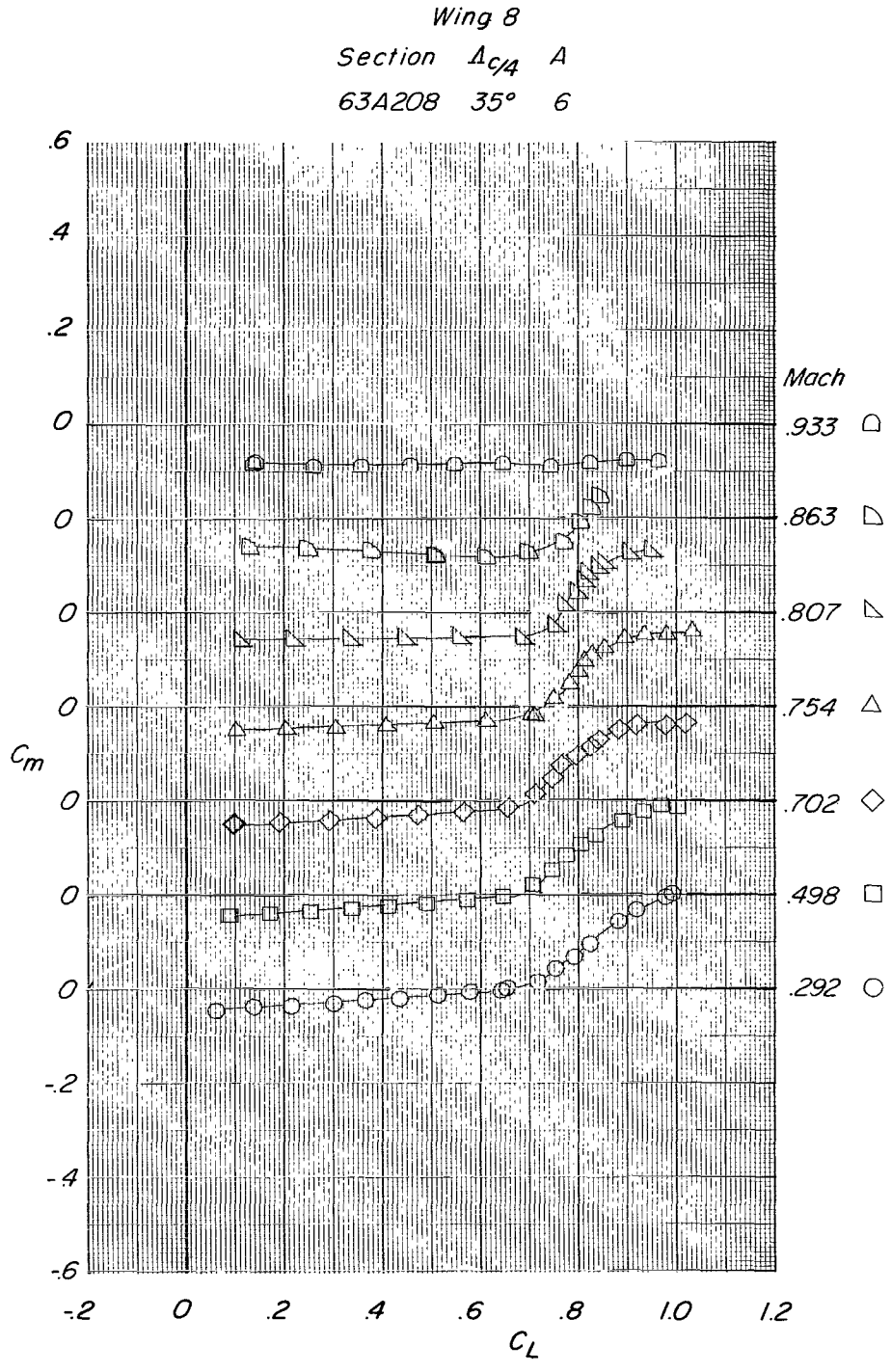
(b) C_A versus α.

Figure 12.- Continued.



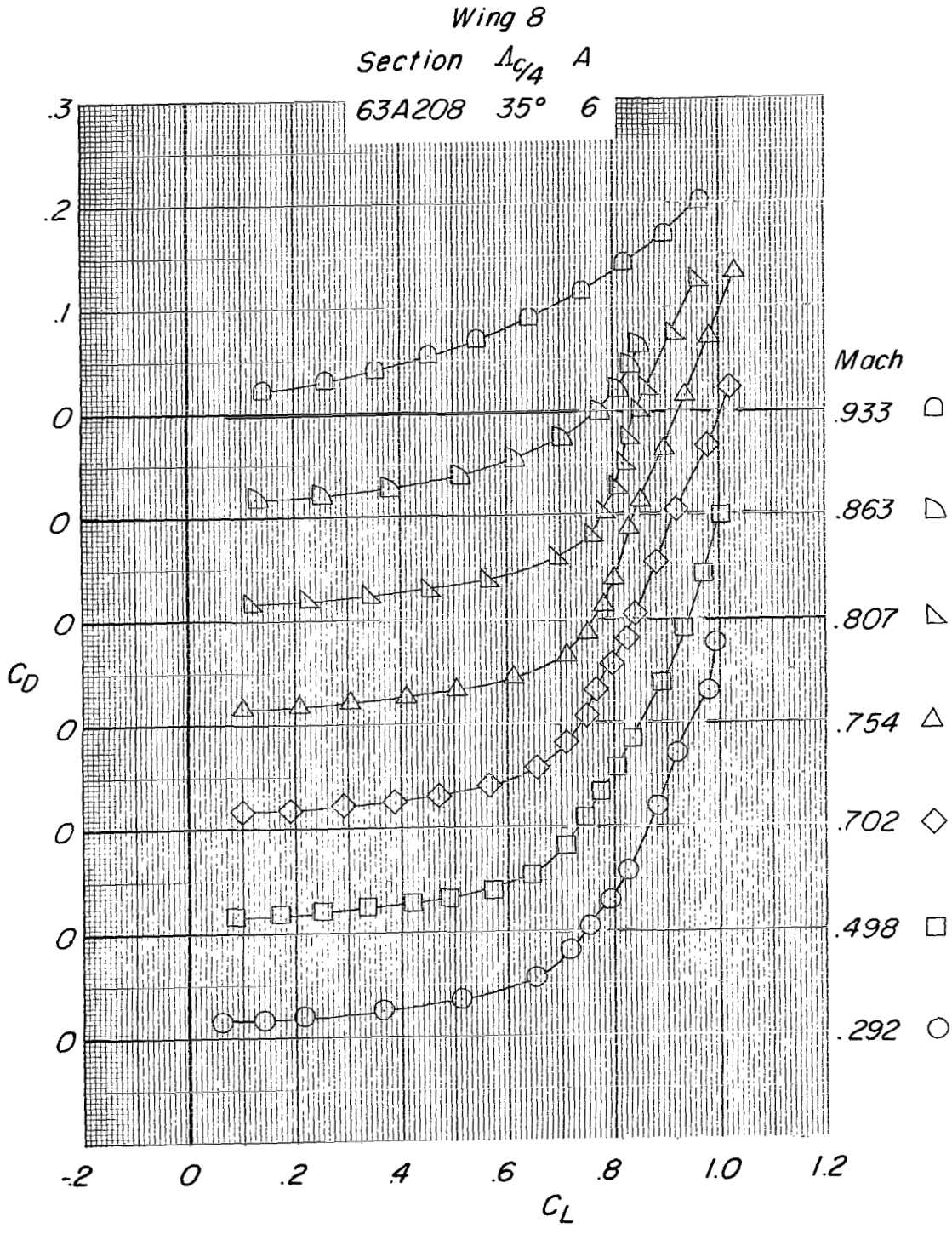
(c) α versus C_L.

Figure 12.- Continued.



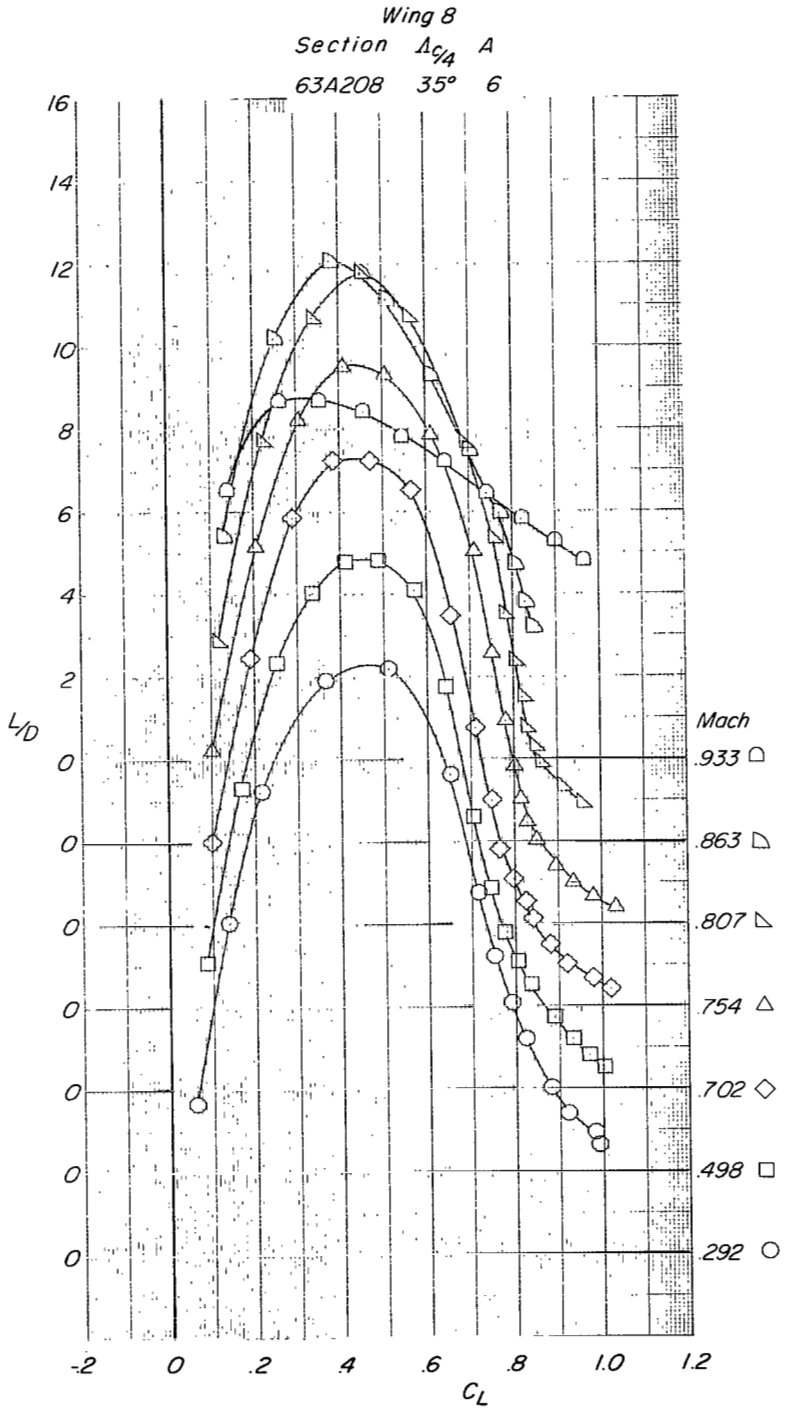
(d) C_m versus C_L.

Figure 12.- Continued.



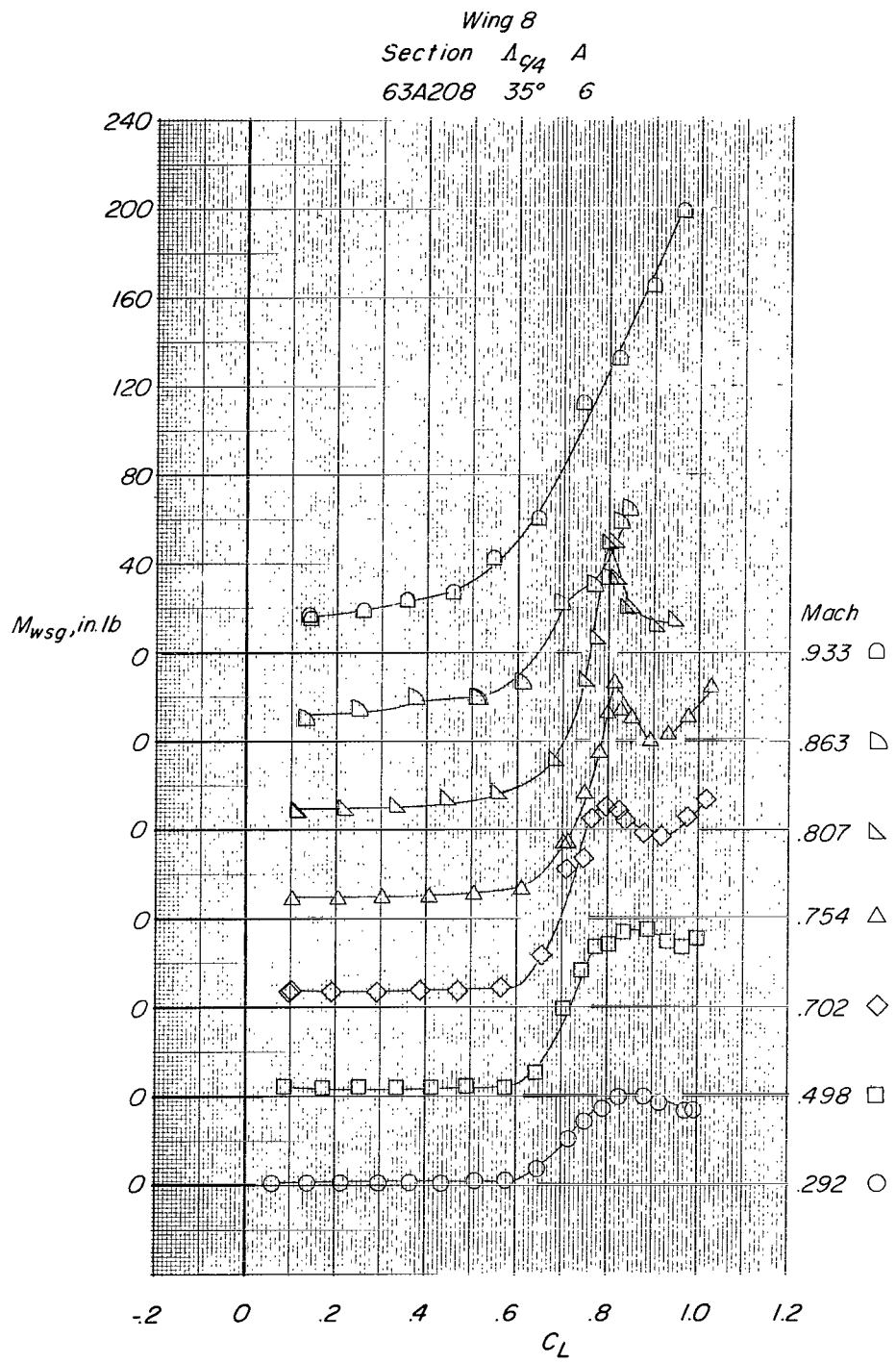
(e) C_D versus C_L.

Figure 12.- Continued.



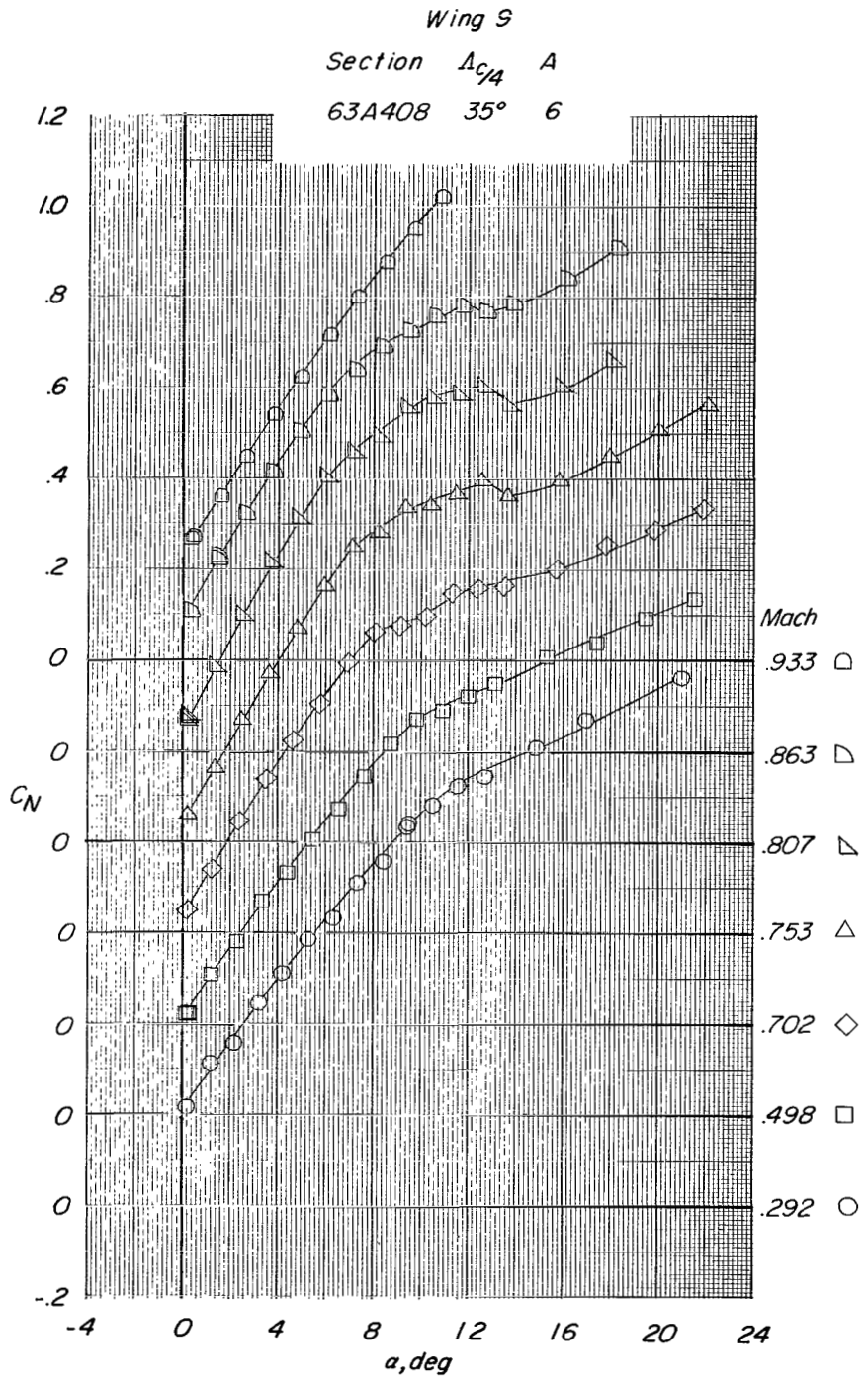
(f) L/D versus C_L.

Figure 12.- Continued.



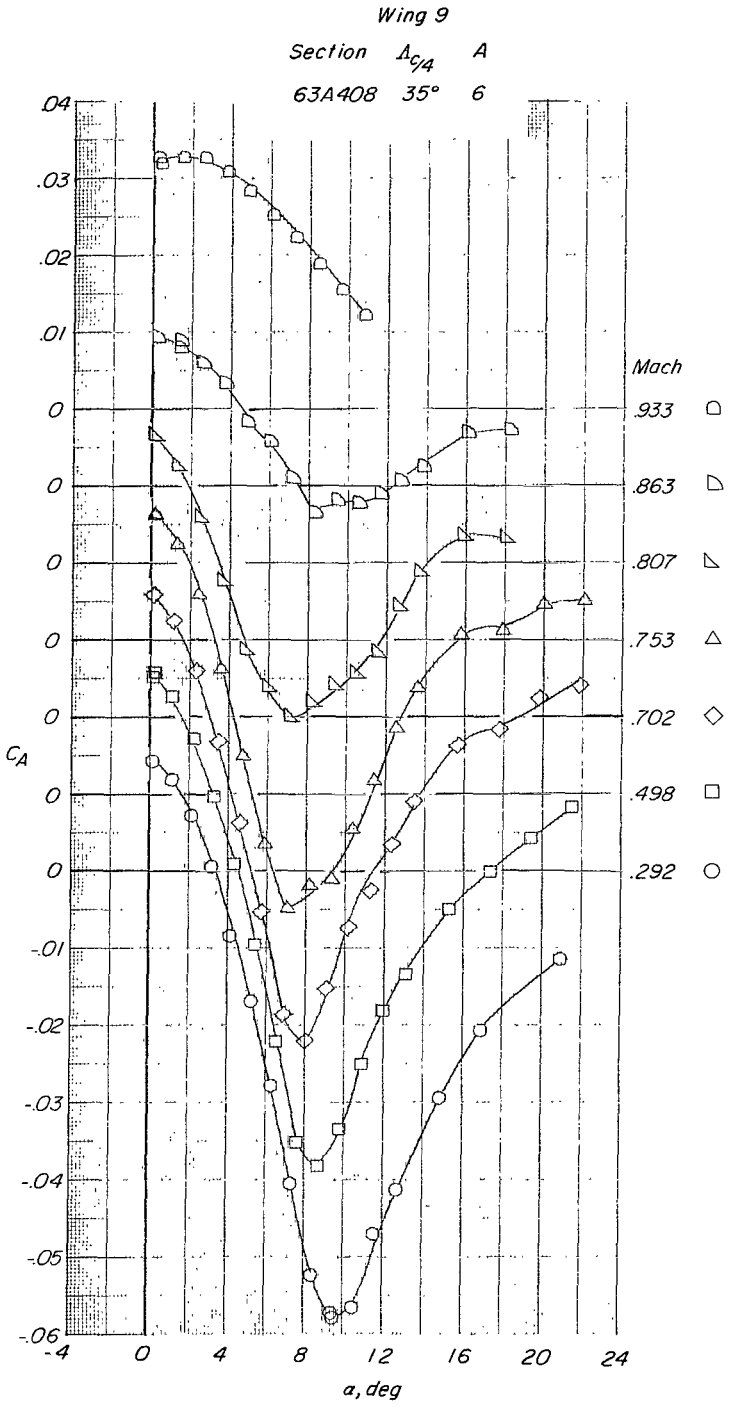
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 12.- Concluded.



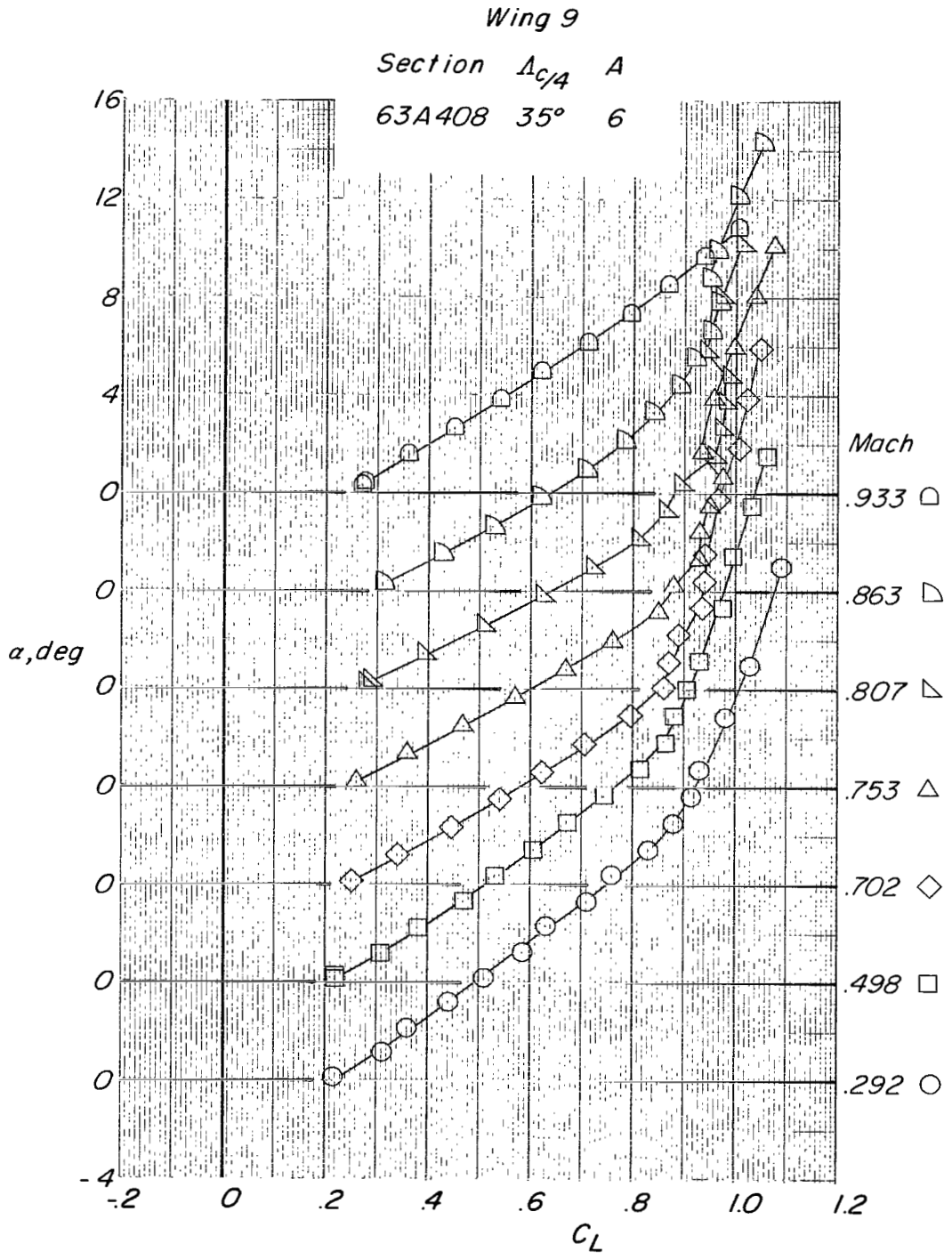
(a) C_N versus α.

Figure 13.- Static longitudinal aerodynamic and buffet characteristics of the wing 9 configuration at Mach numbers from 0.29 to 0.93. Pointed forebody; transition grit off.



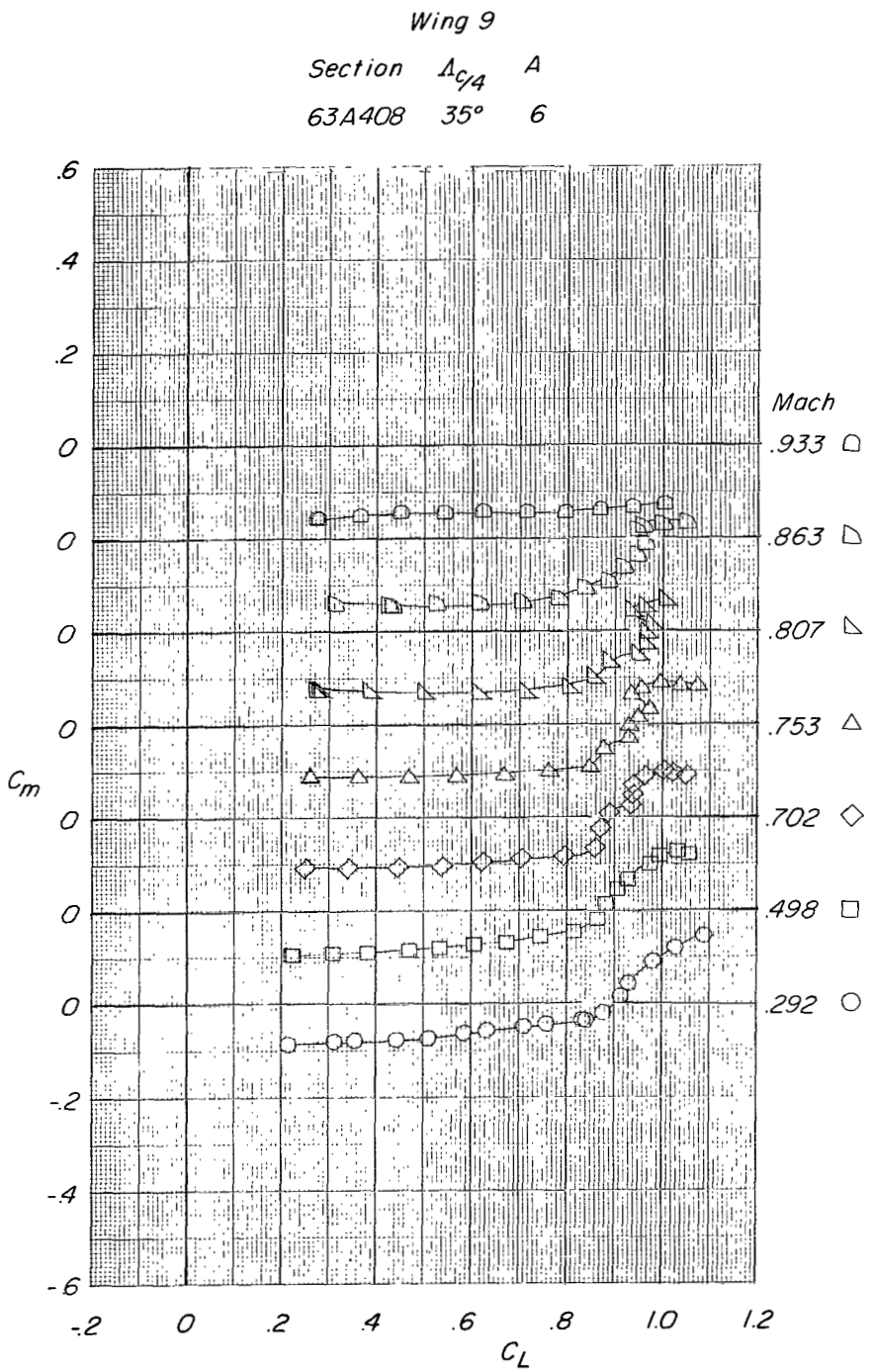
(b) C_A versus α.

Figure 13.- Continued.



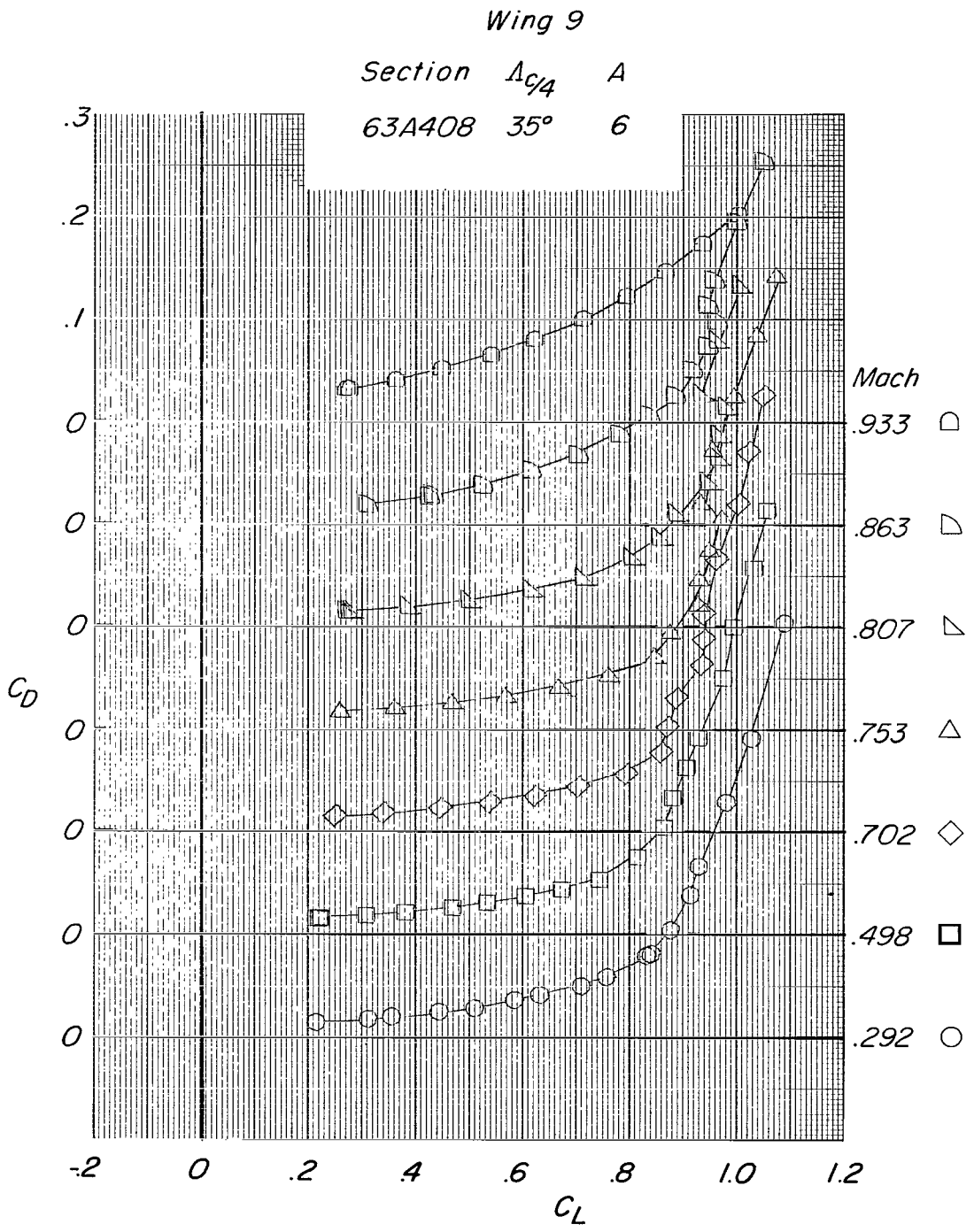
(c) α versus C_L.

Figure 13.- Continued.



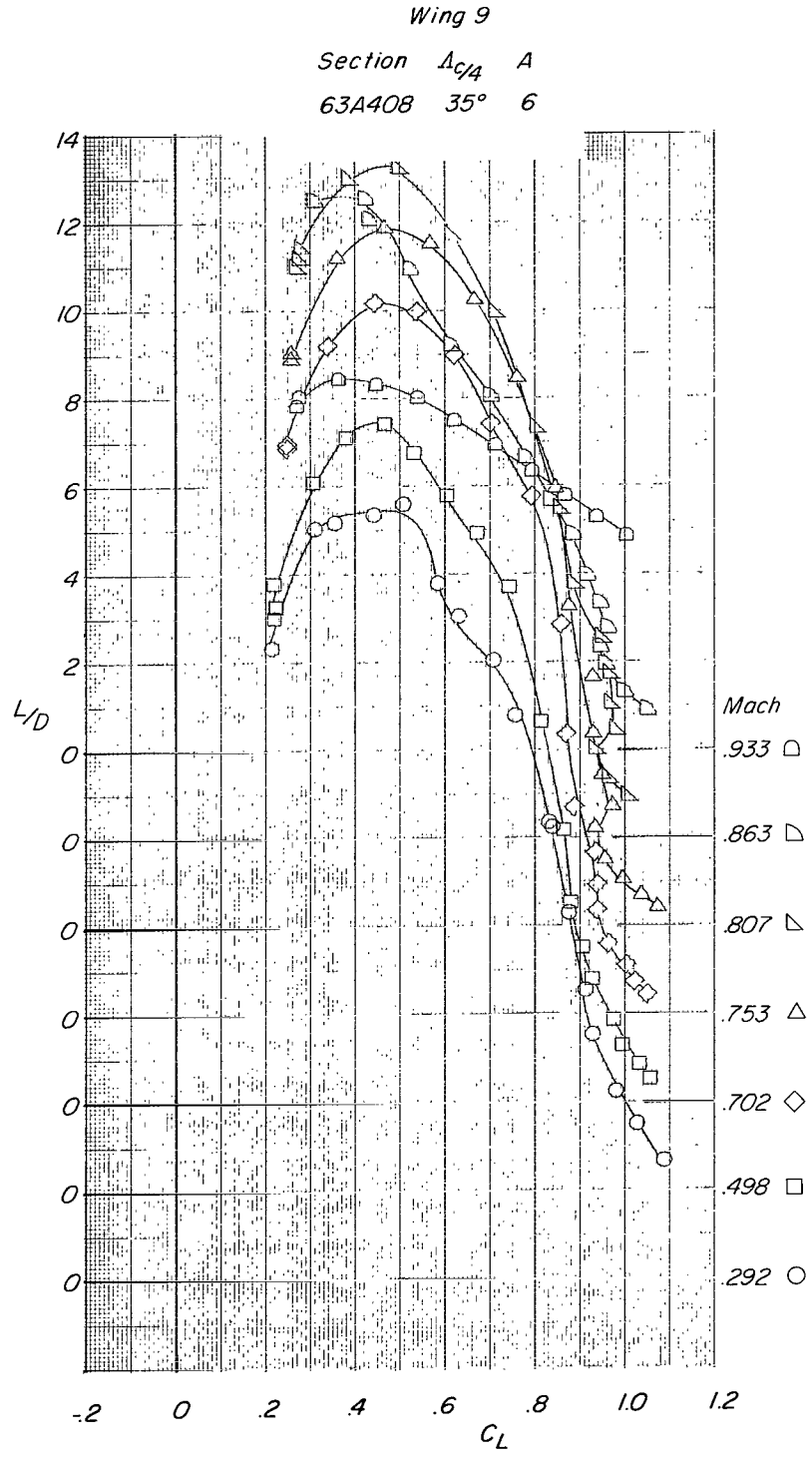
(d) C_m versus C_L.

Figure 13.- Continued.



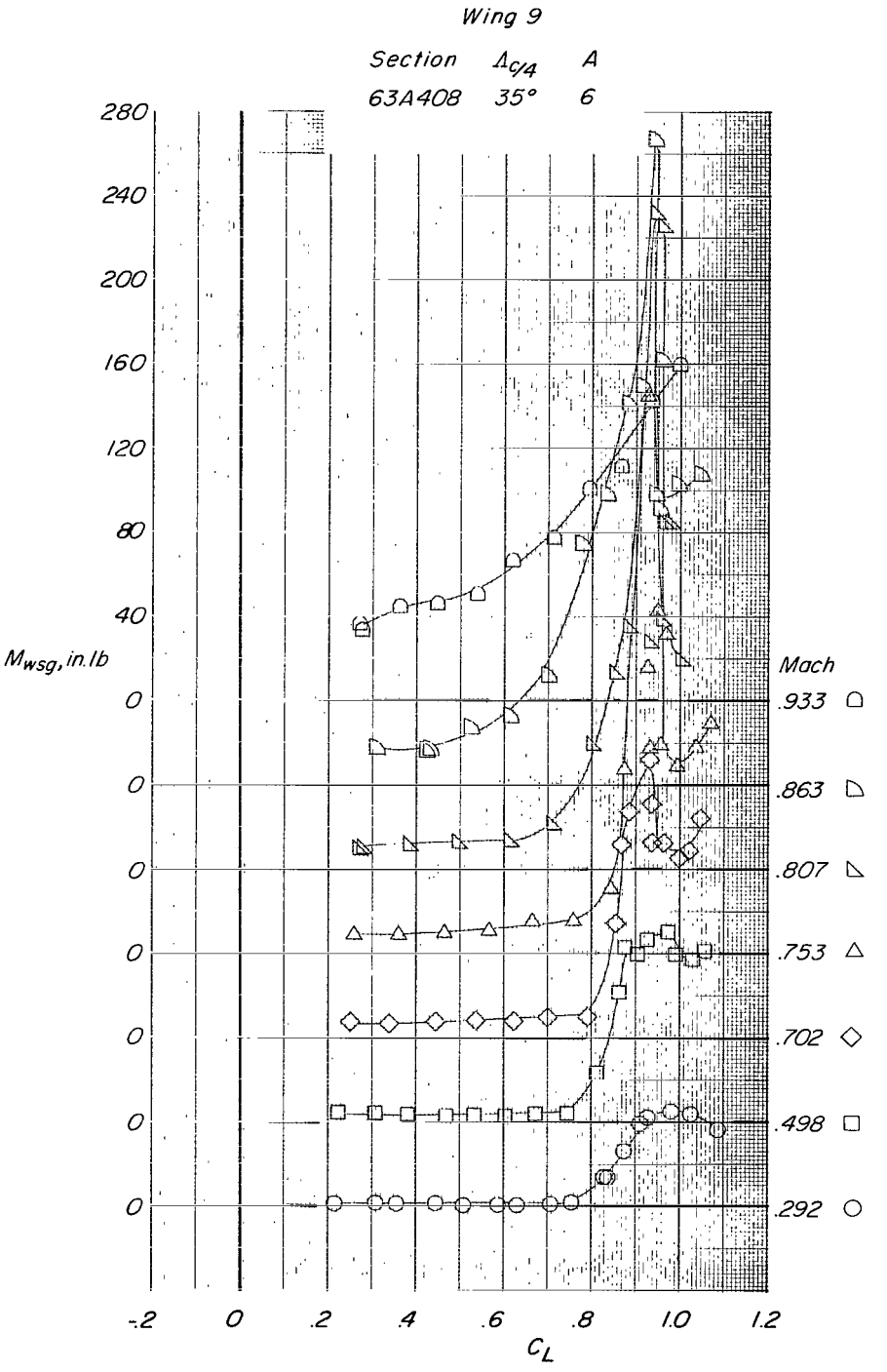
(e) C_D versus C_L.

Figure 13.- Continued.



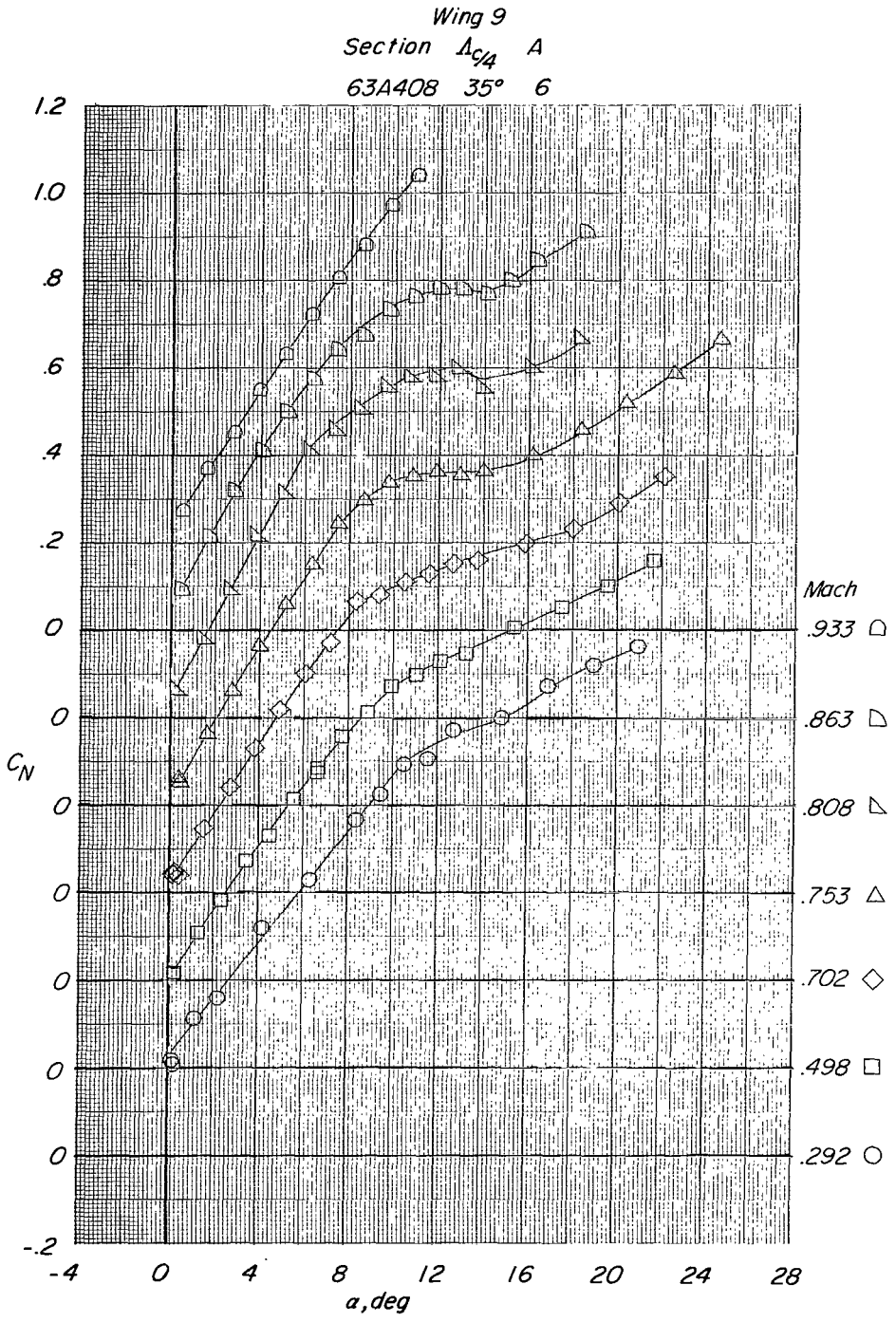
(f) L/D versus C_L.

Figure 13.- Continued.



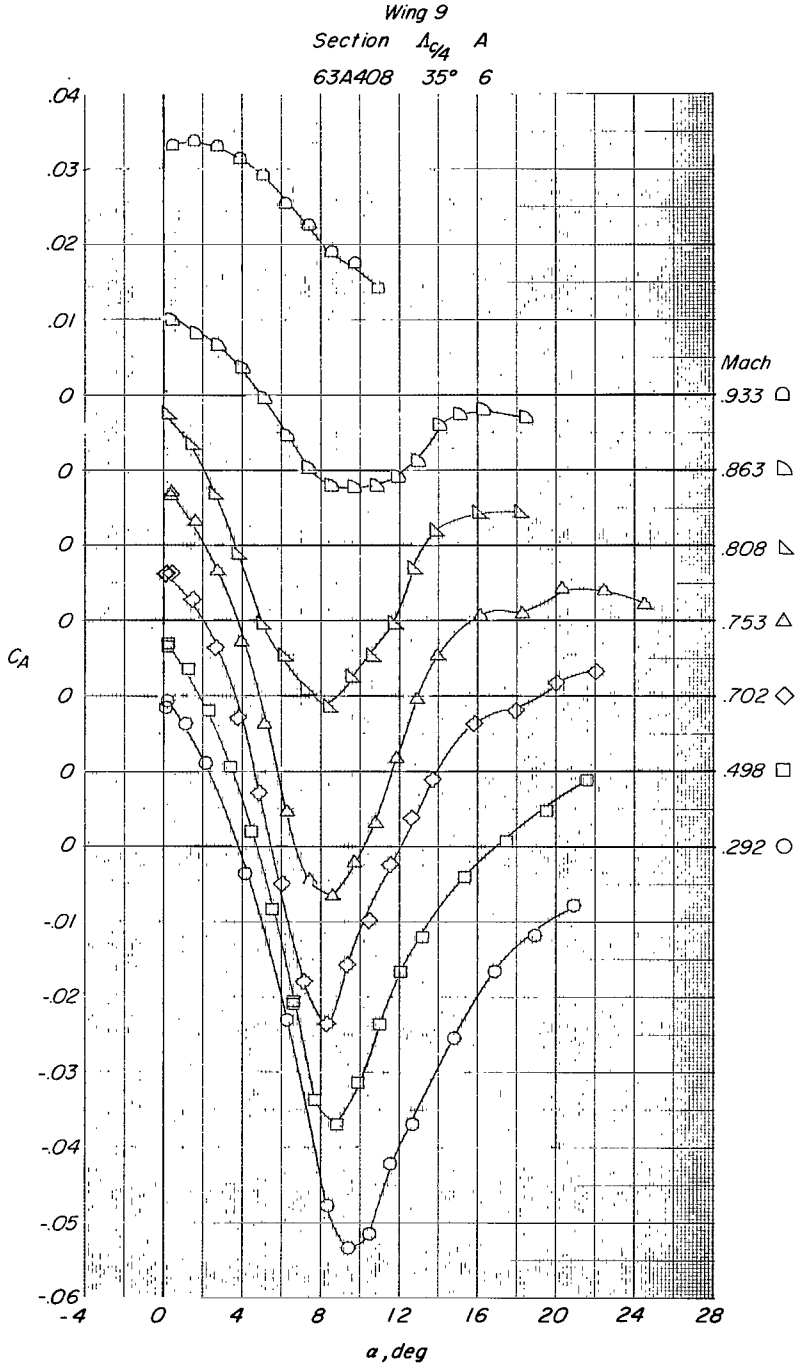
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 13.- Concluded.



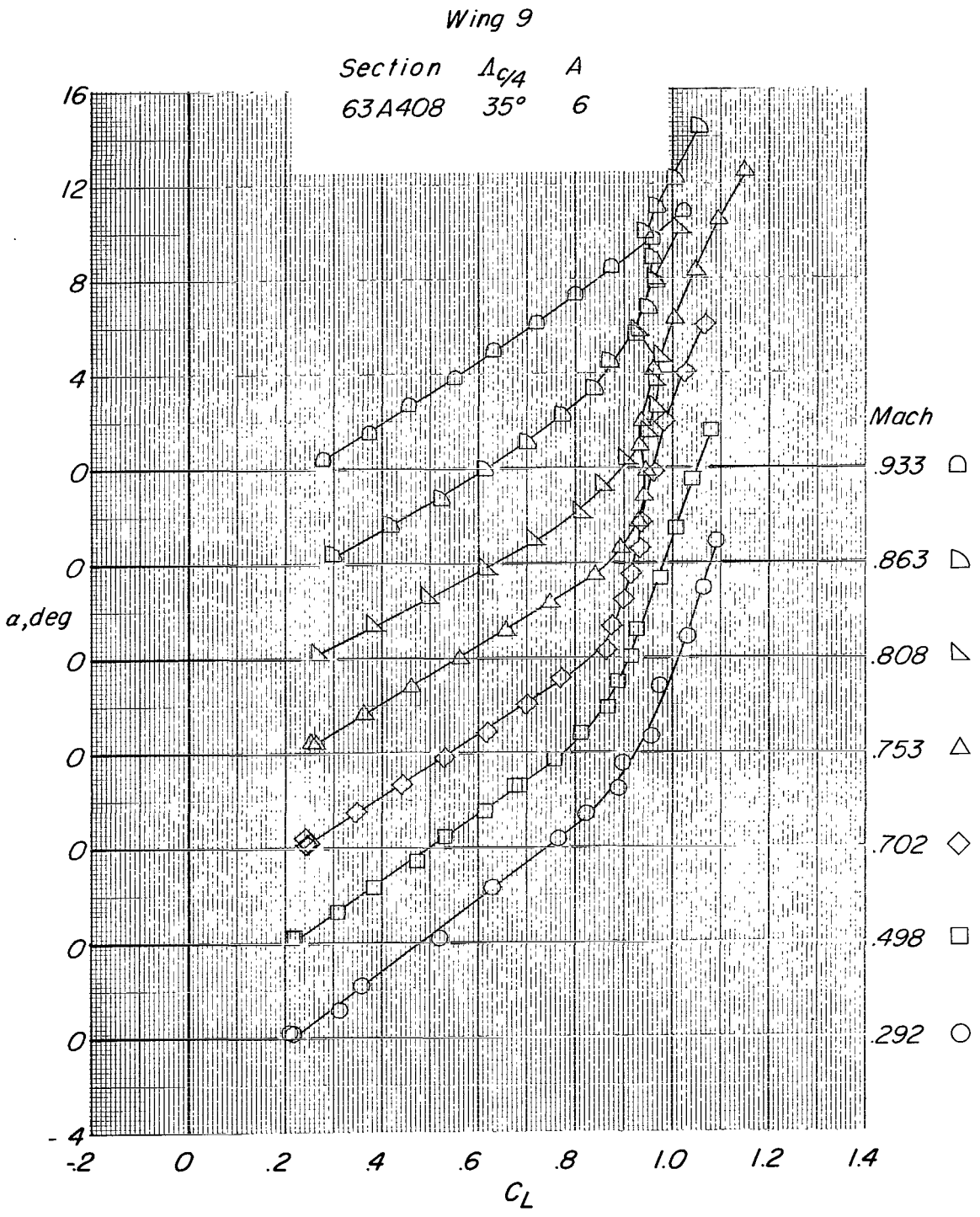
(a) C_N versus α.

Figure 14.- Static longitudinal aerodynamic and buffet characteristics of the wing 9 configuration at Mach numbers from 0.29 to 0.93. Rounded forebody; transition grit off.



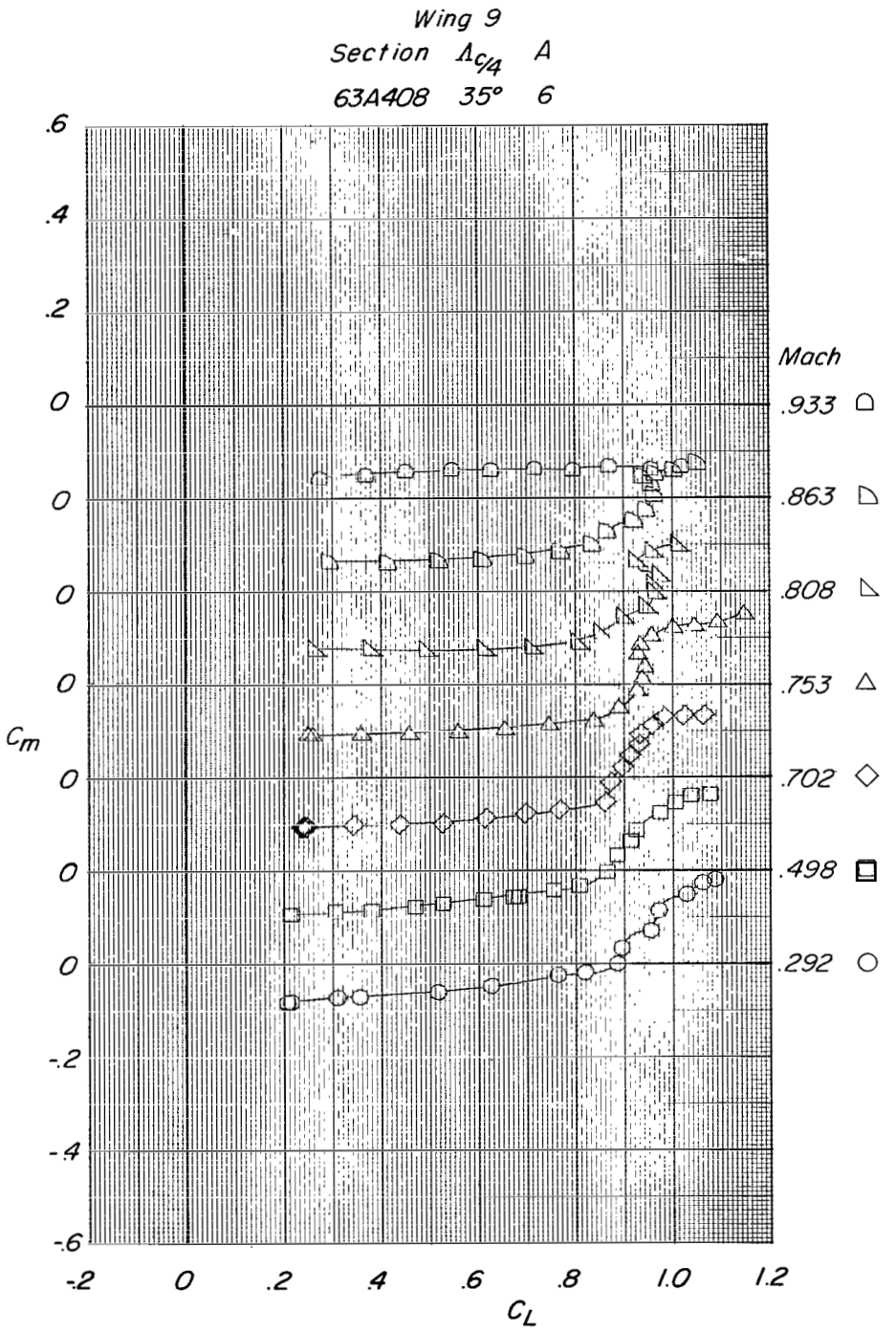
(b) C_A versus α.

Figure 14.- Continued.



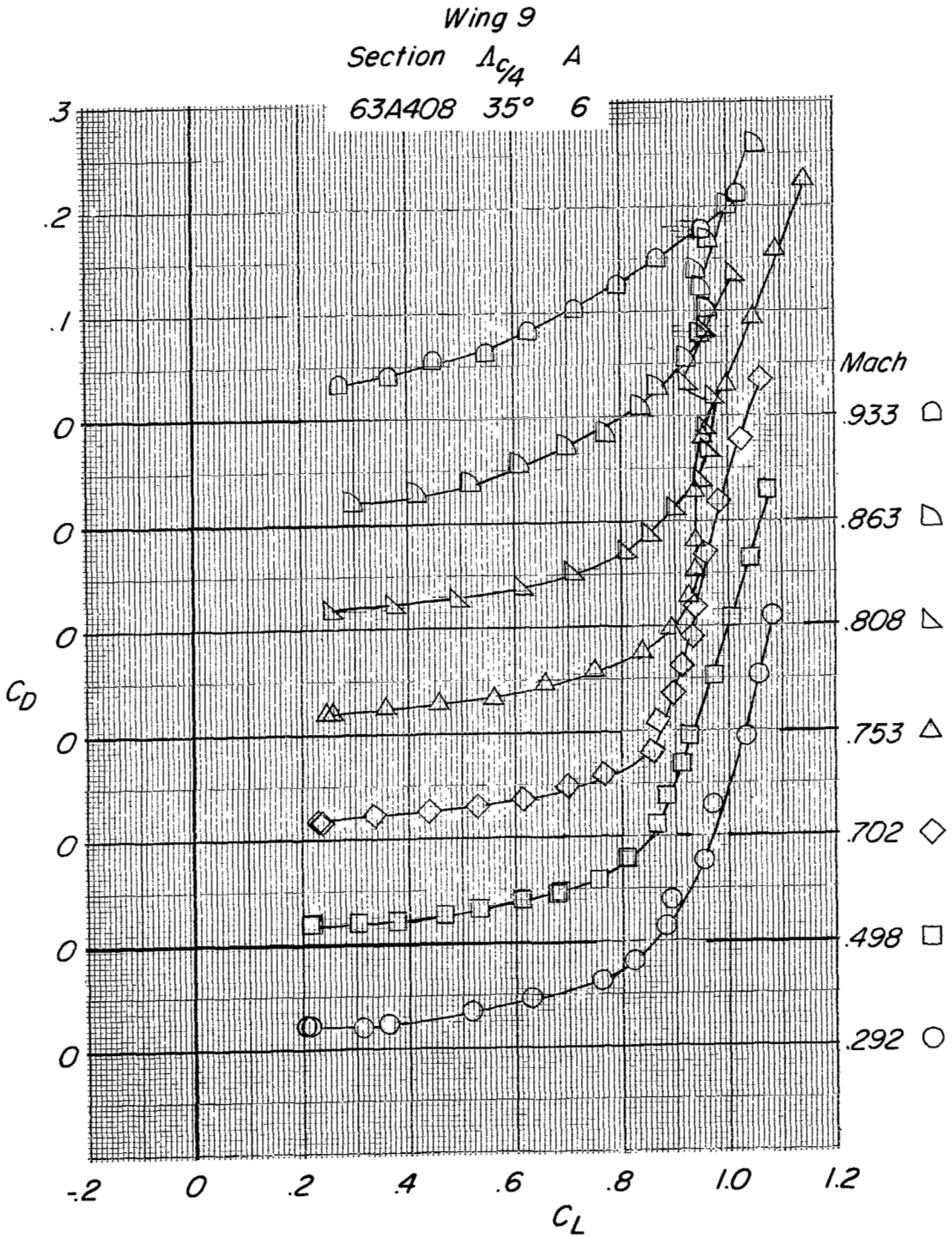
(c) α versus C_L.

Figure 14.- Continued.



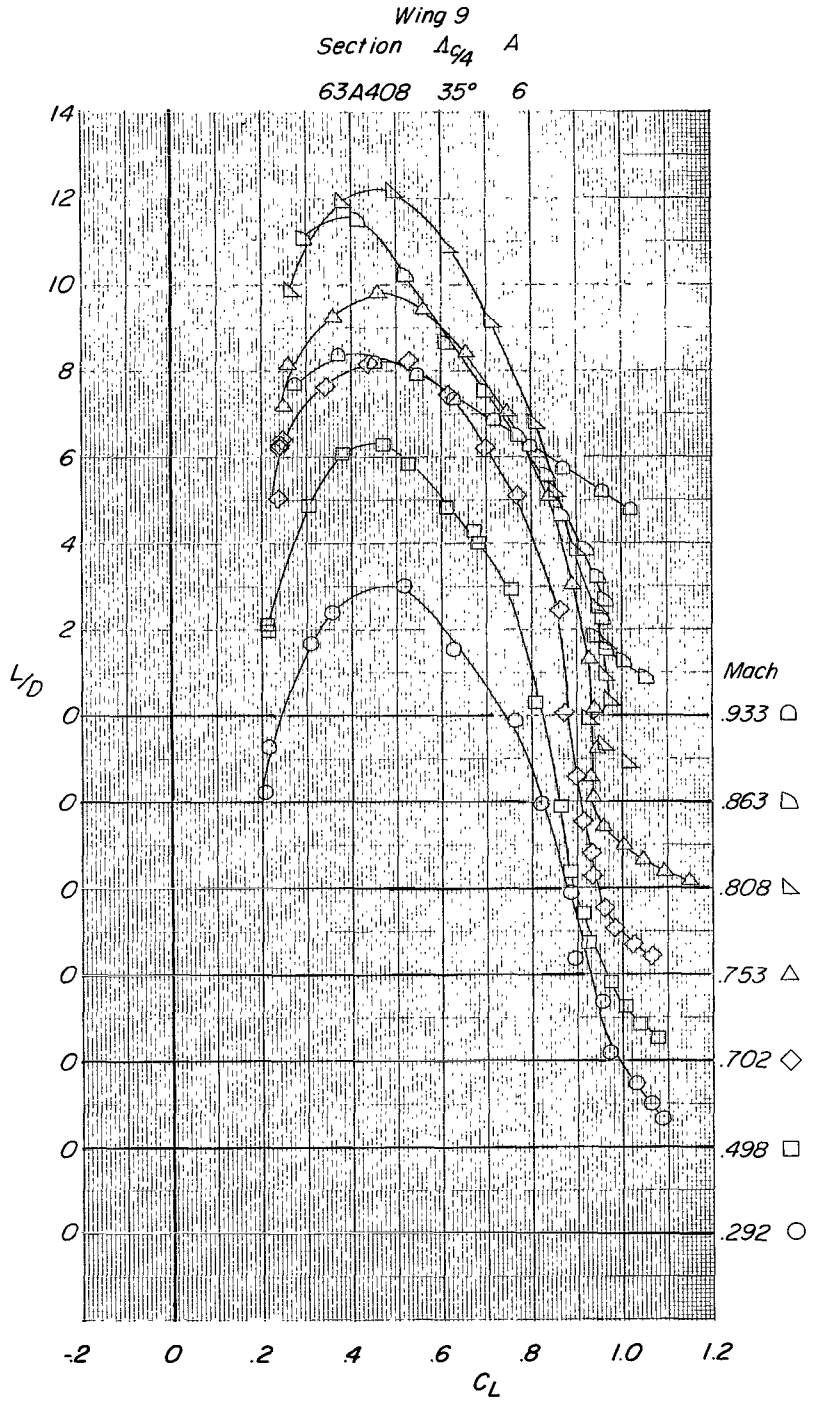
(d) C_m versus C_L.

Figure 14.- Continued.



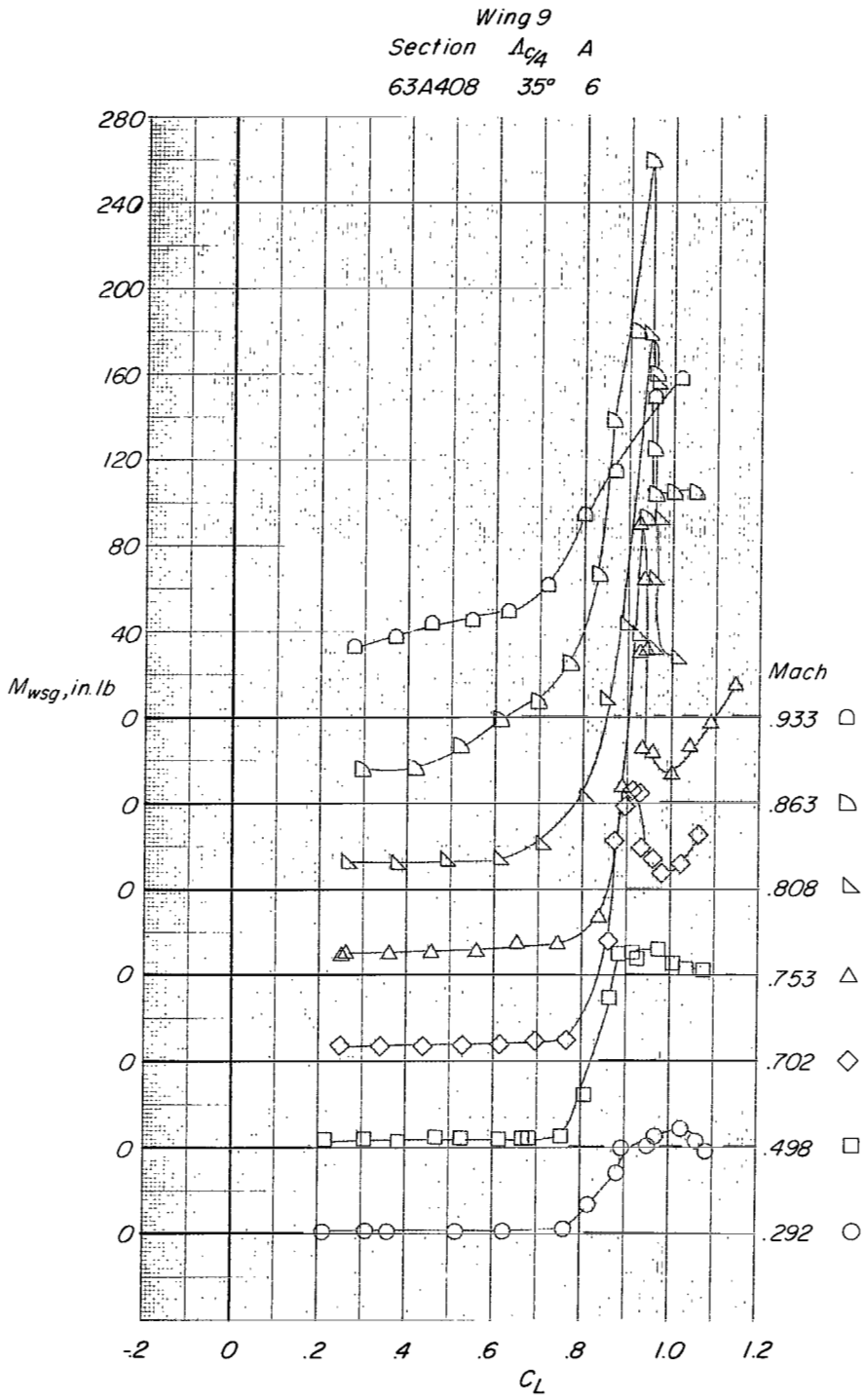
(e) C_D versus C_L.

Figure 14.- Continued.



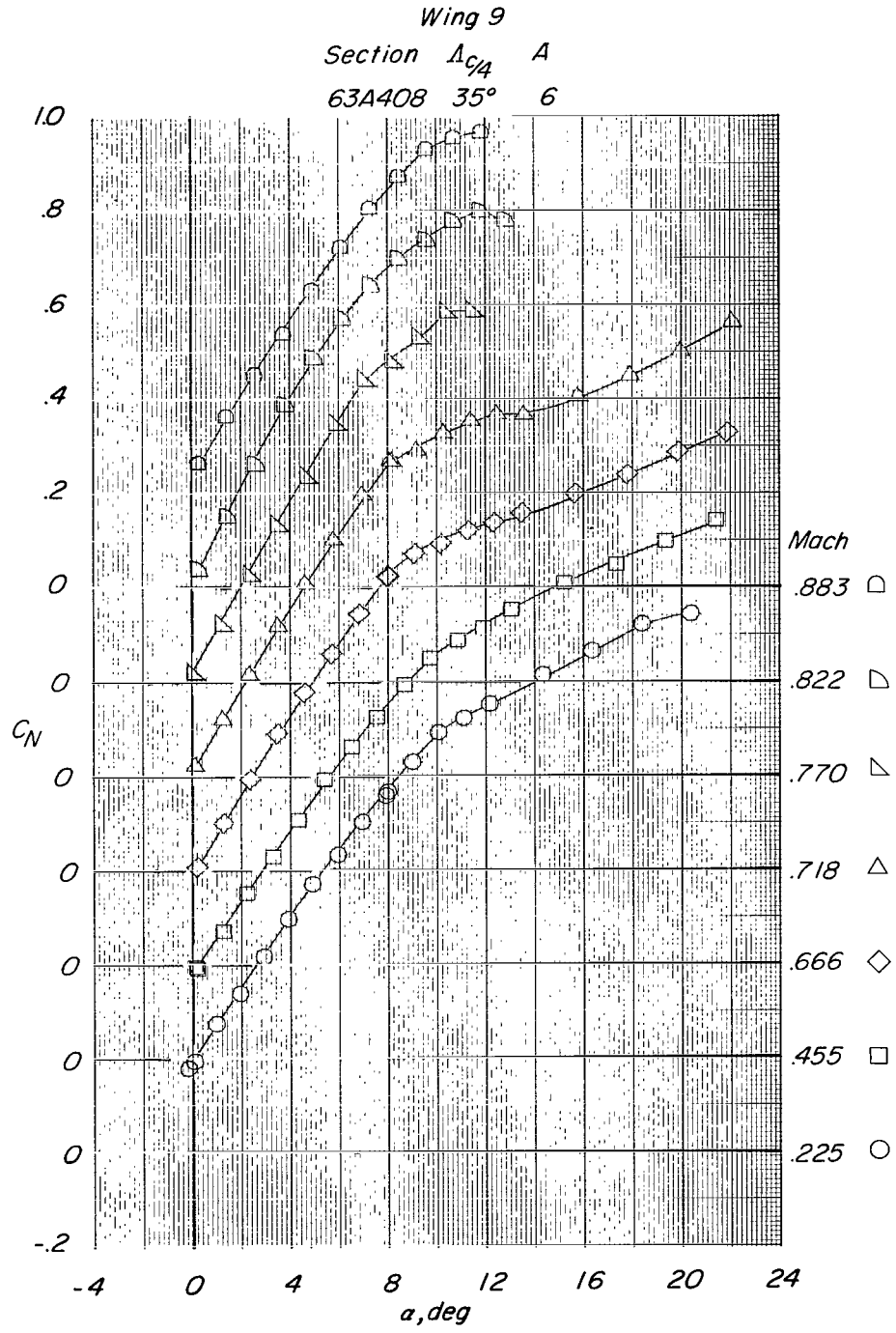
(f) L/D versus C_L.

Figure 14.- Continued.



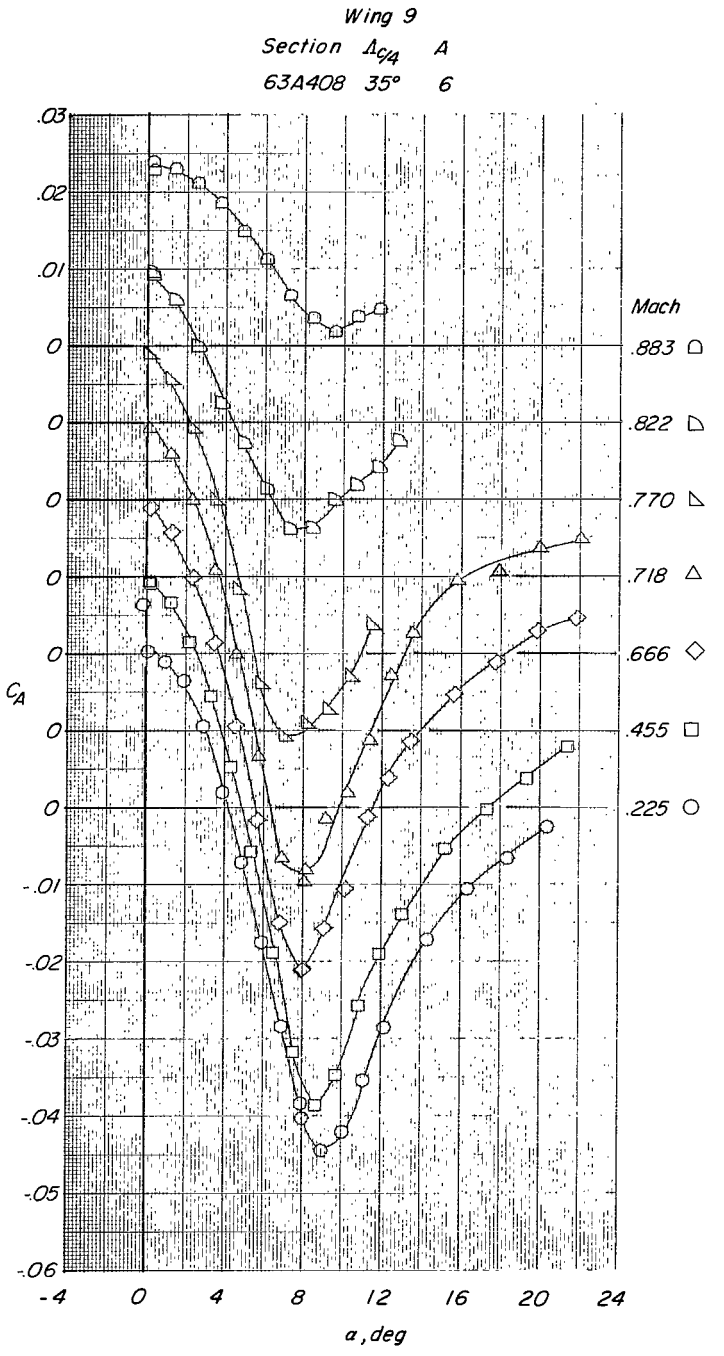
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 14.- Concluded.



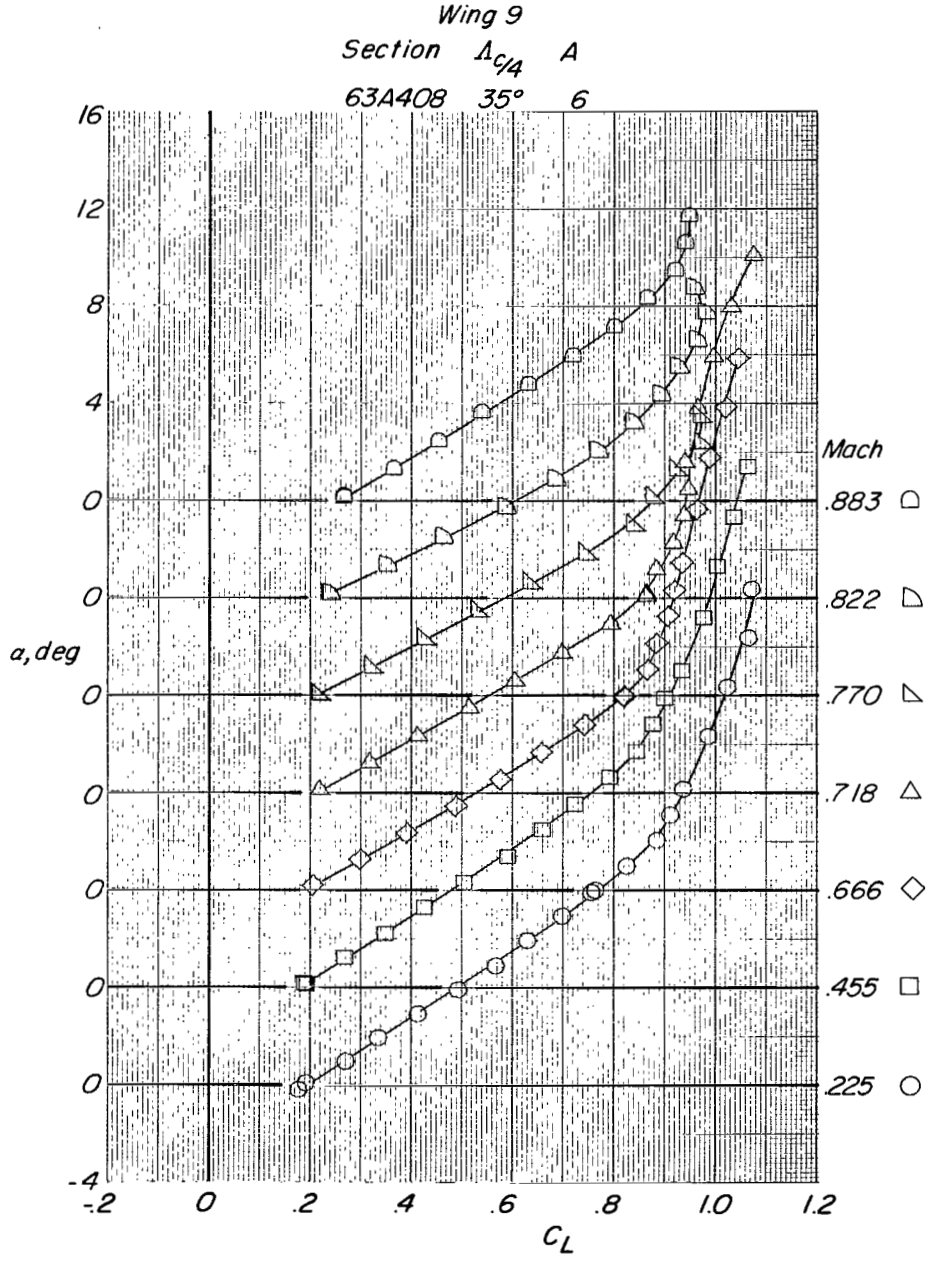
(a) C_N versus α.

Figure 15.- Static longitudinal aerodynamic and buffet characteristics of the wing 9 configuration at Mach numbers from 0.23 to 0.88. Rounded forebody; transition grit on.



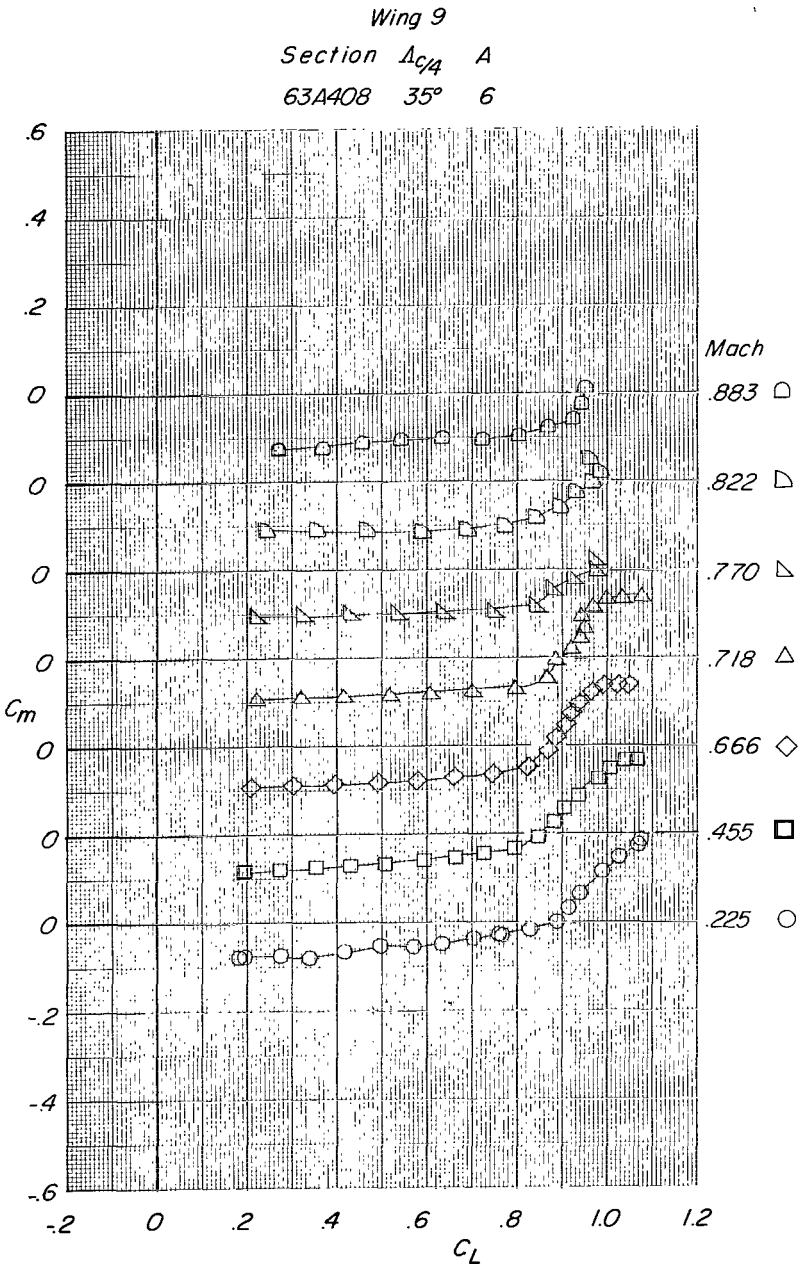
(b) C_A versus α.

Figure 15.- Continued.



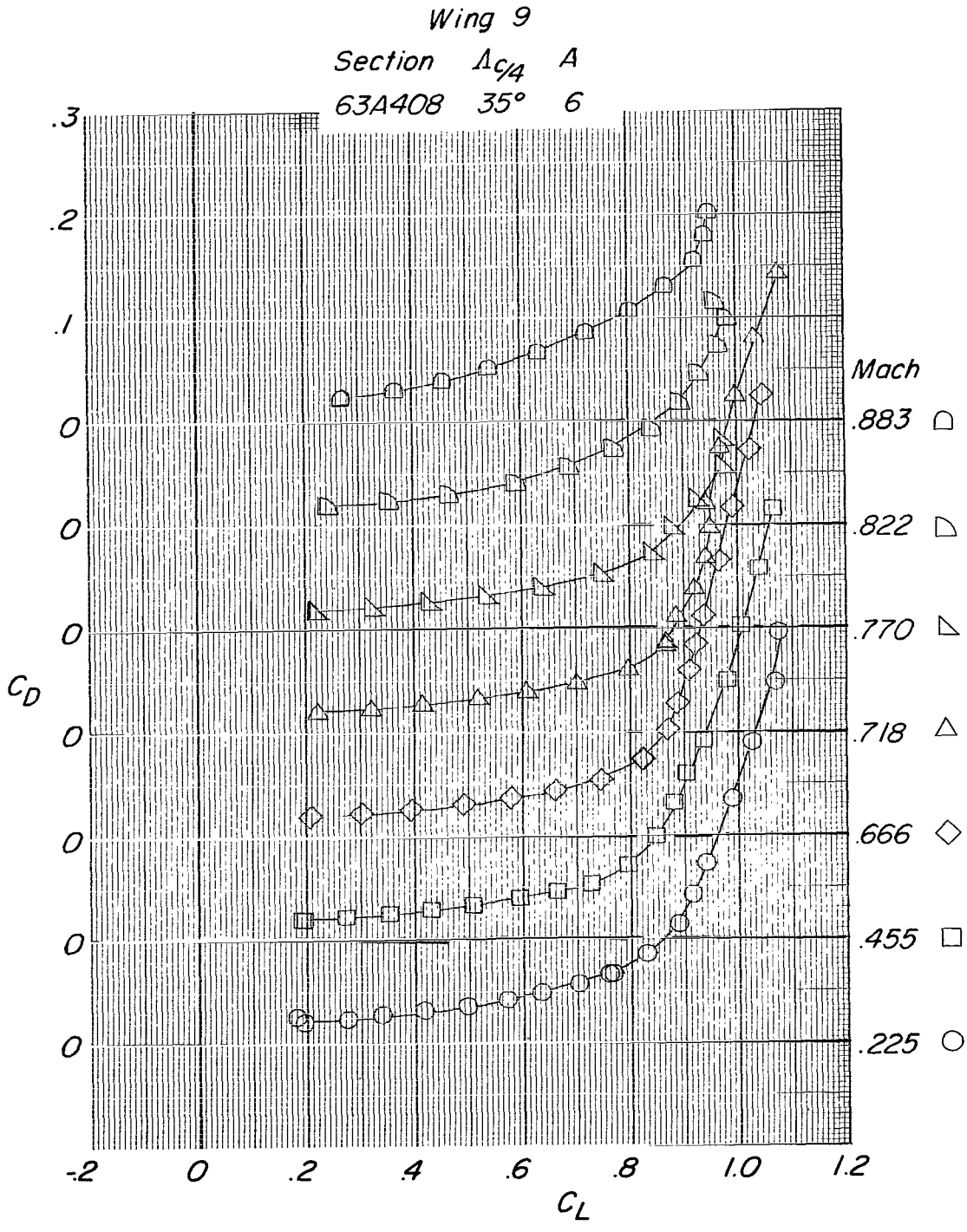
(c) α versus C_L.

Figure 15.- Continued.



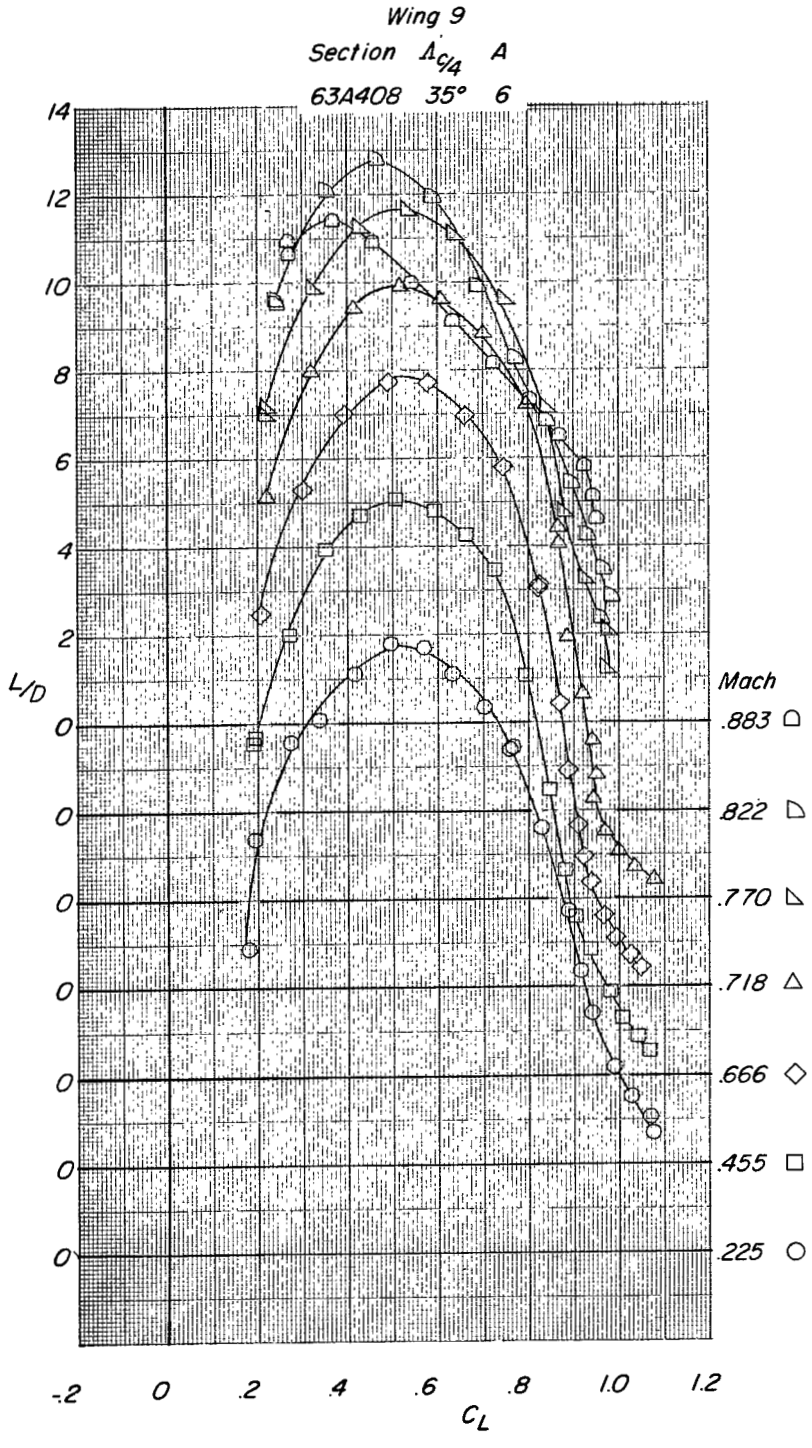
(d) C_m versus C_L.

Figure 15.- Continued.



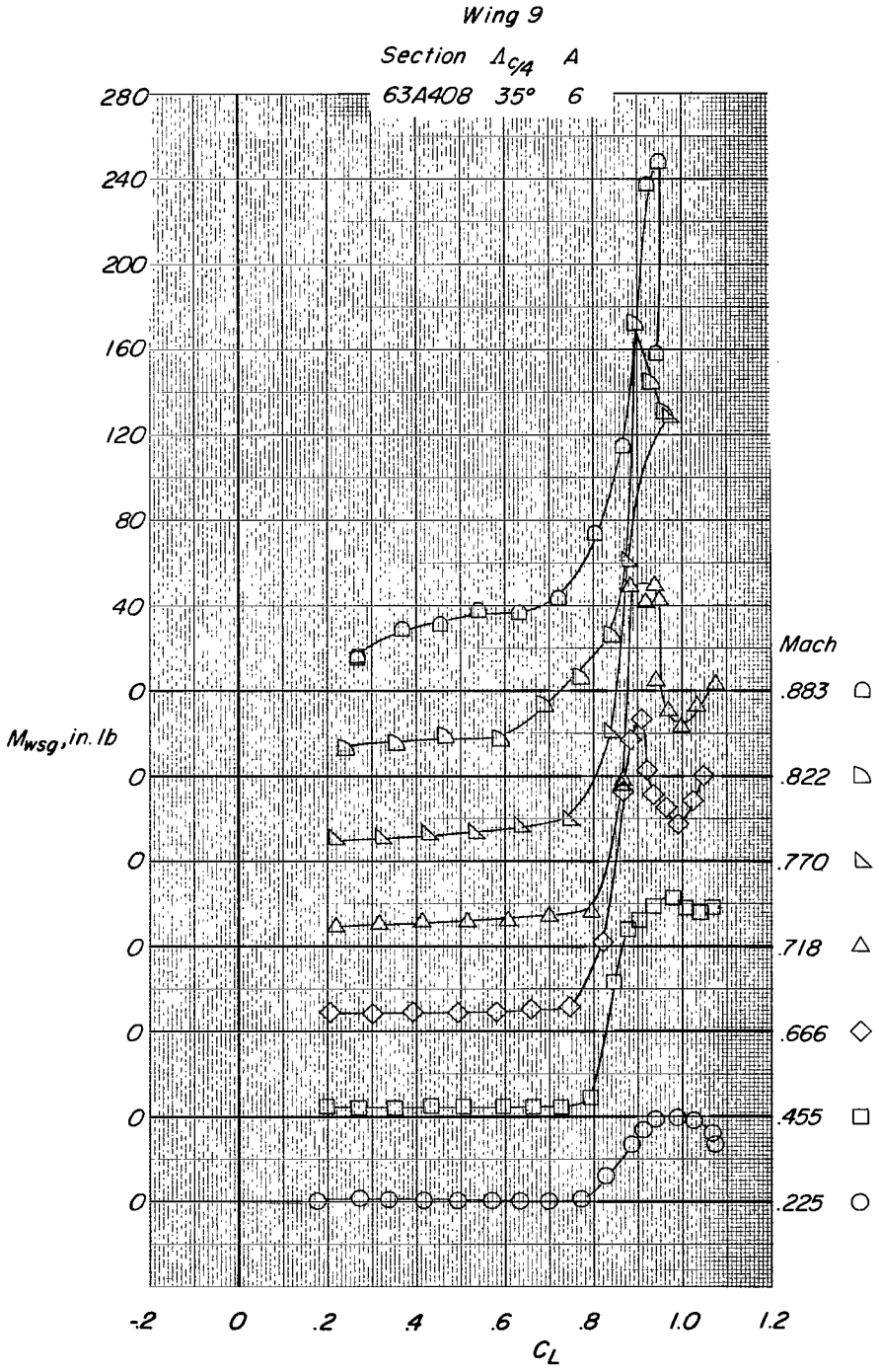
(e) C_D versus C_L.

Figure 15.- Continued.



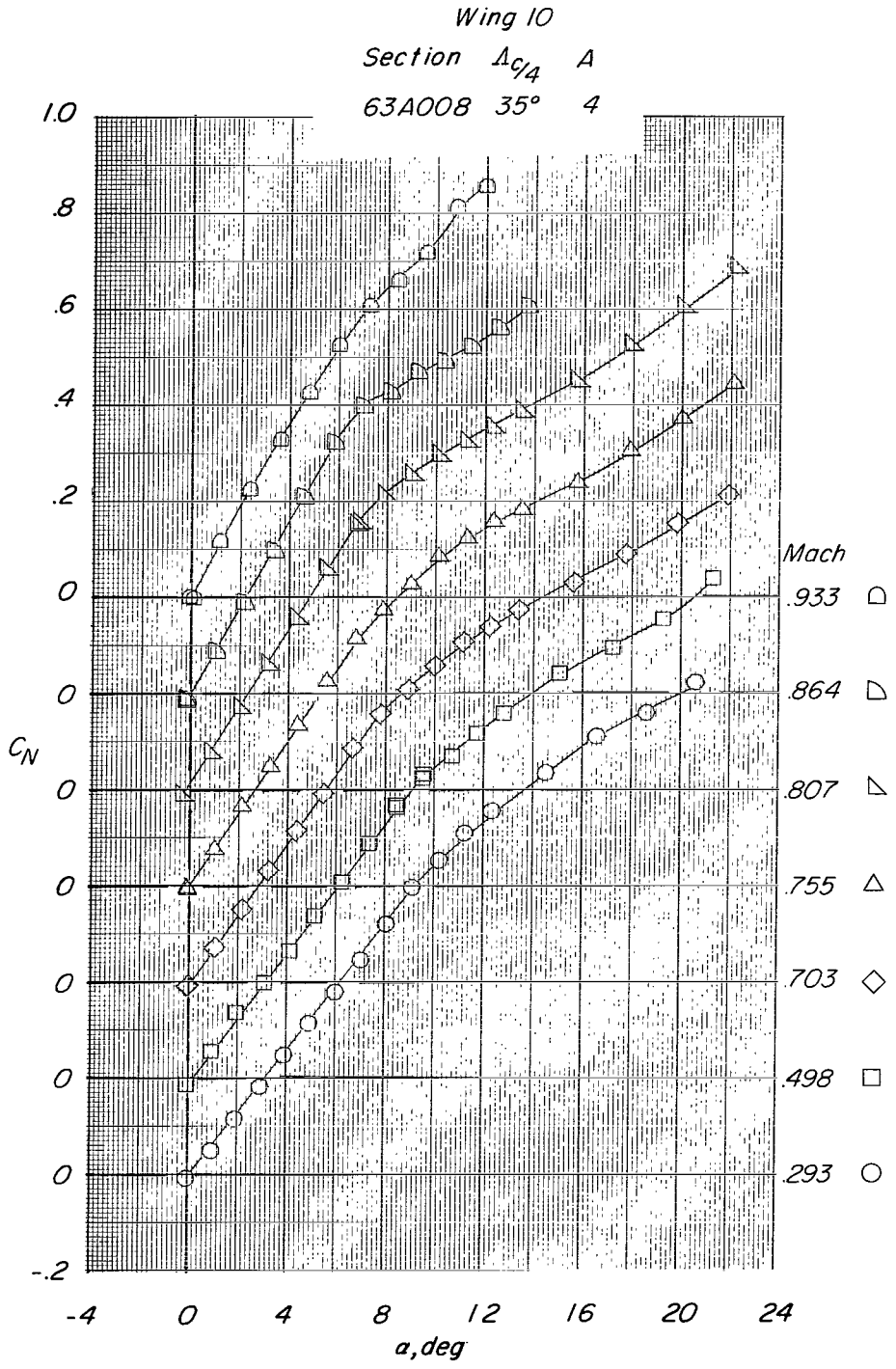
(f) L/D versus C_L.

Figure 15.- Continued.



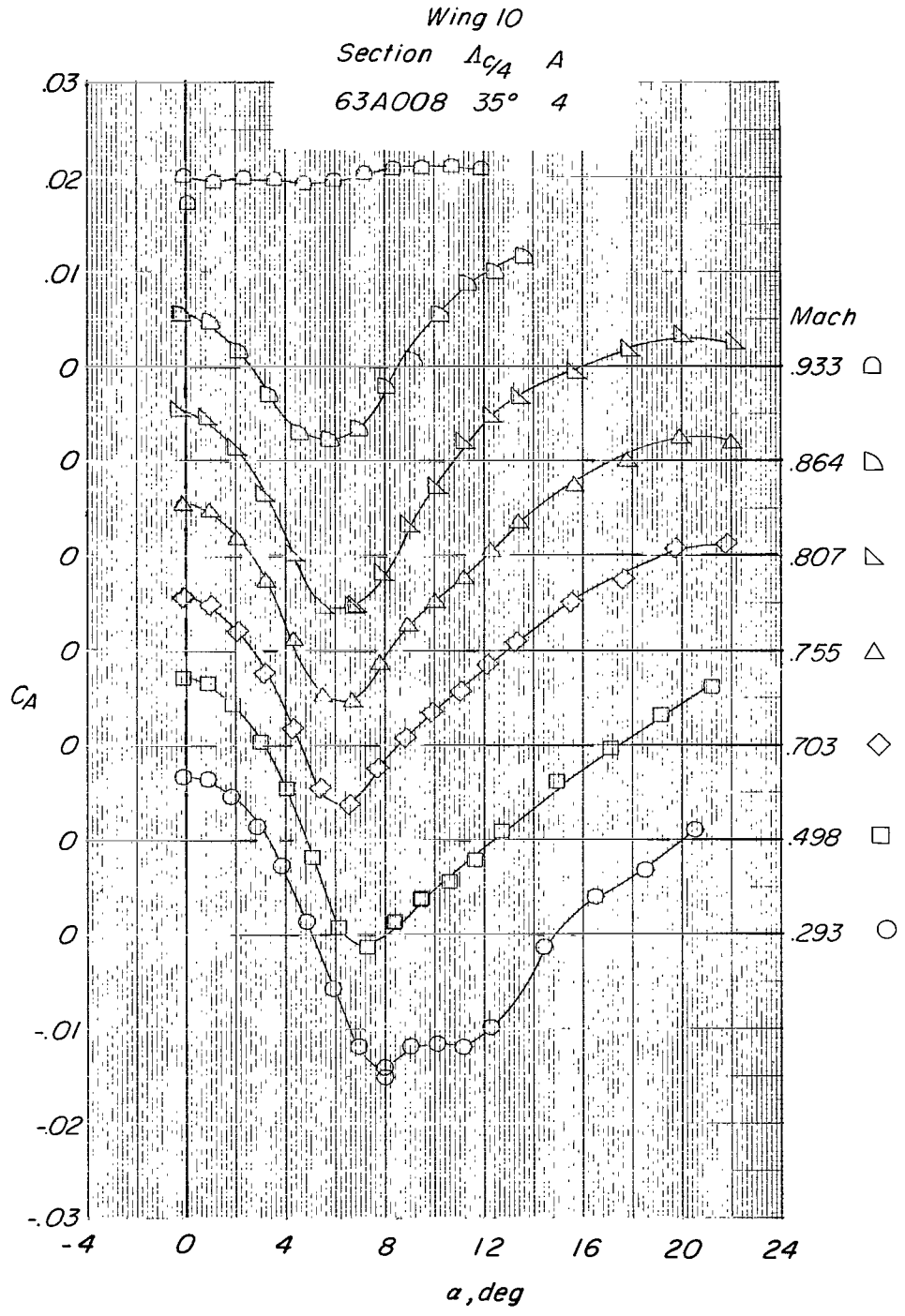
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 15.- Concluded.



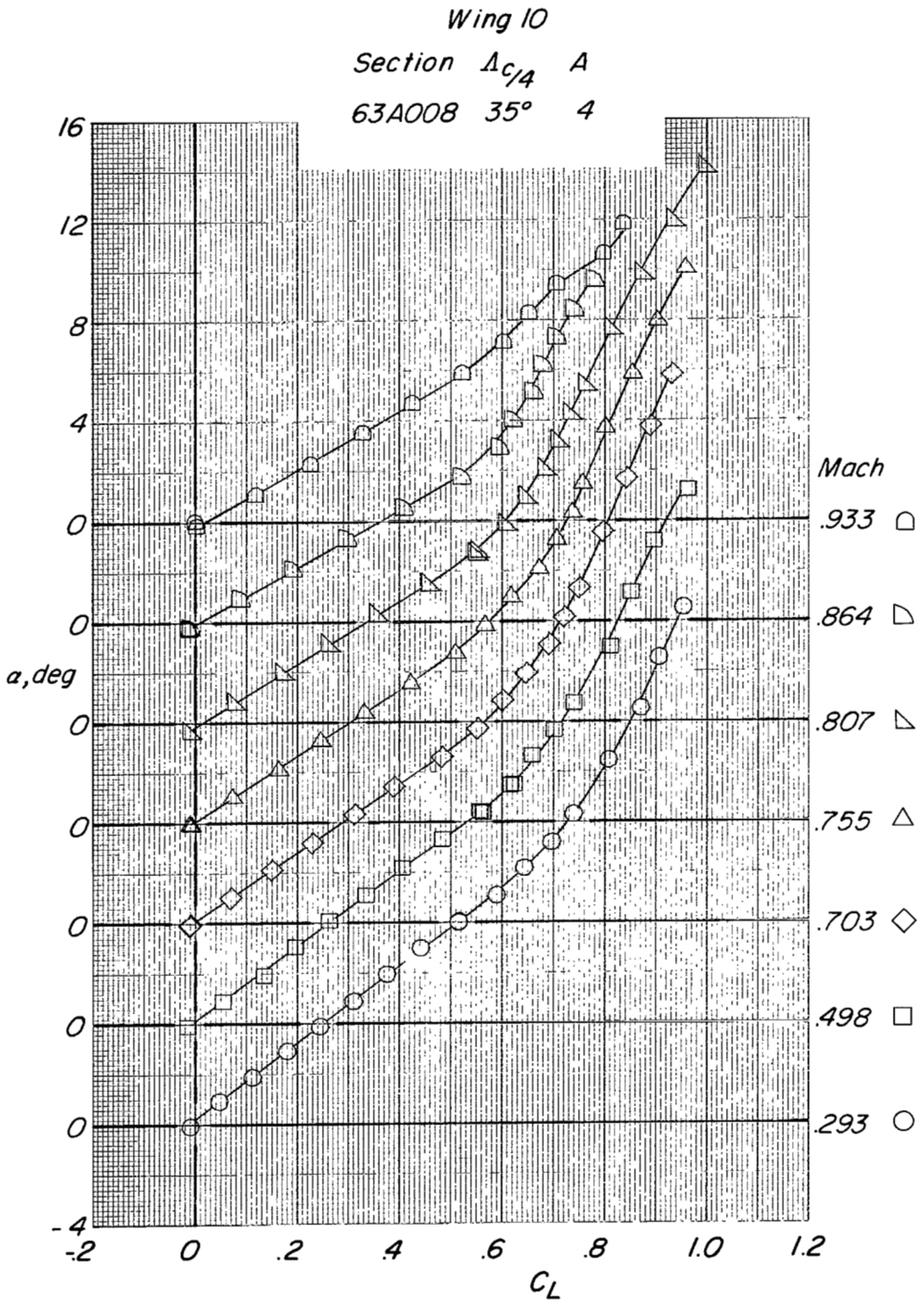
(a) C_N versus α.

Figure 16.- Static longitudinal aerodynamic and buffet characteristics of the wing 10 configuration at Mach numbers from 0.29 to 0.93. Rounded forebody; transition grit on.



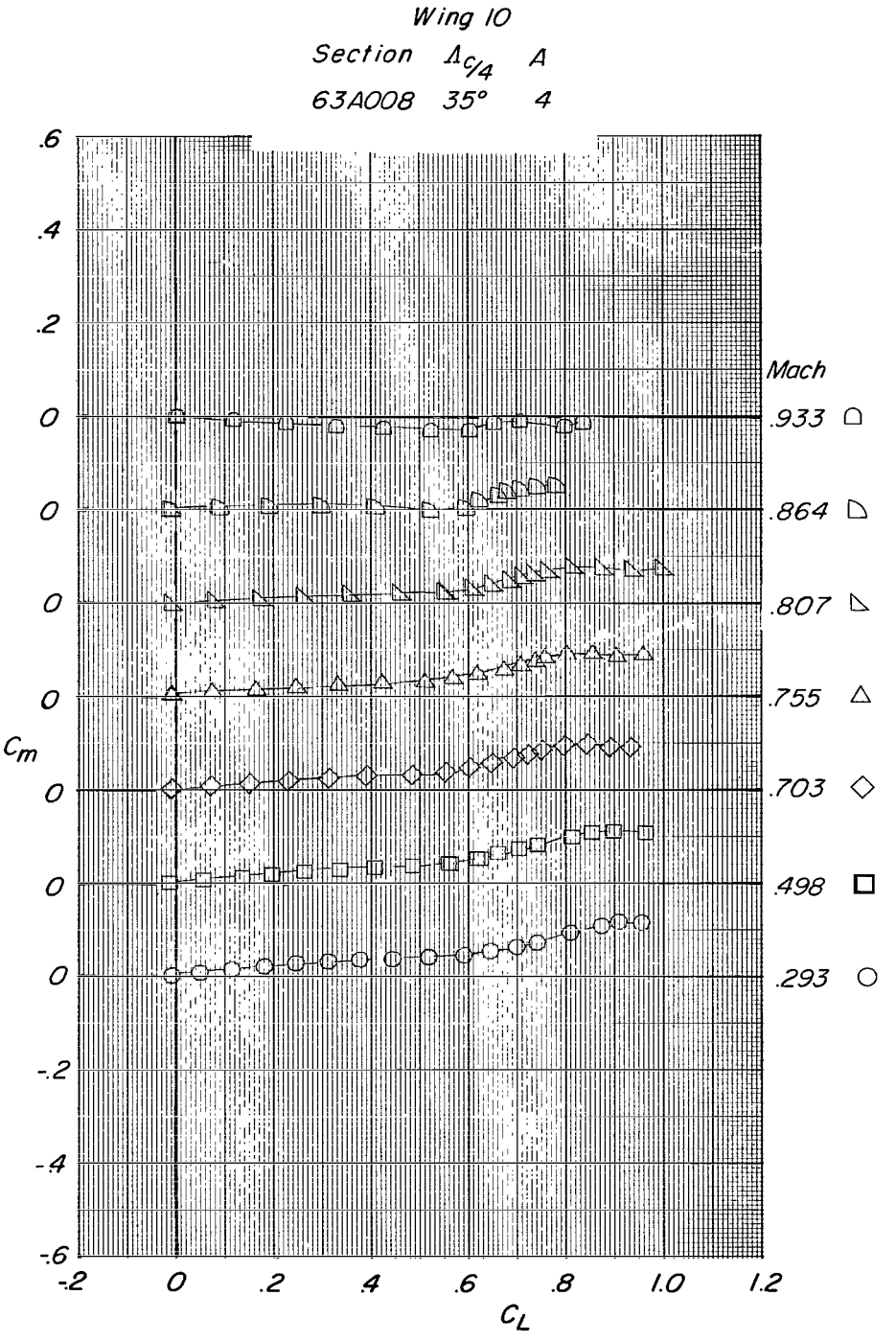
(b) C_A versus α.

Figure 16.- Continued.



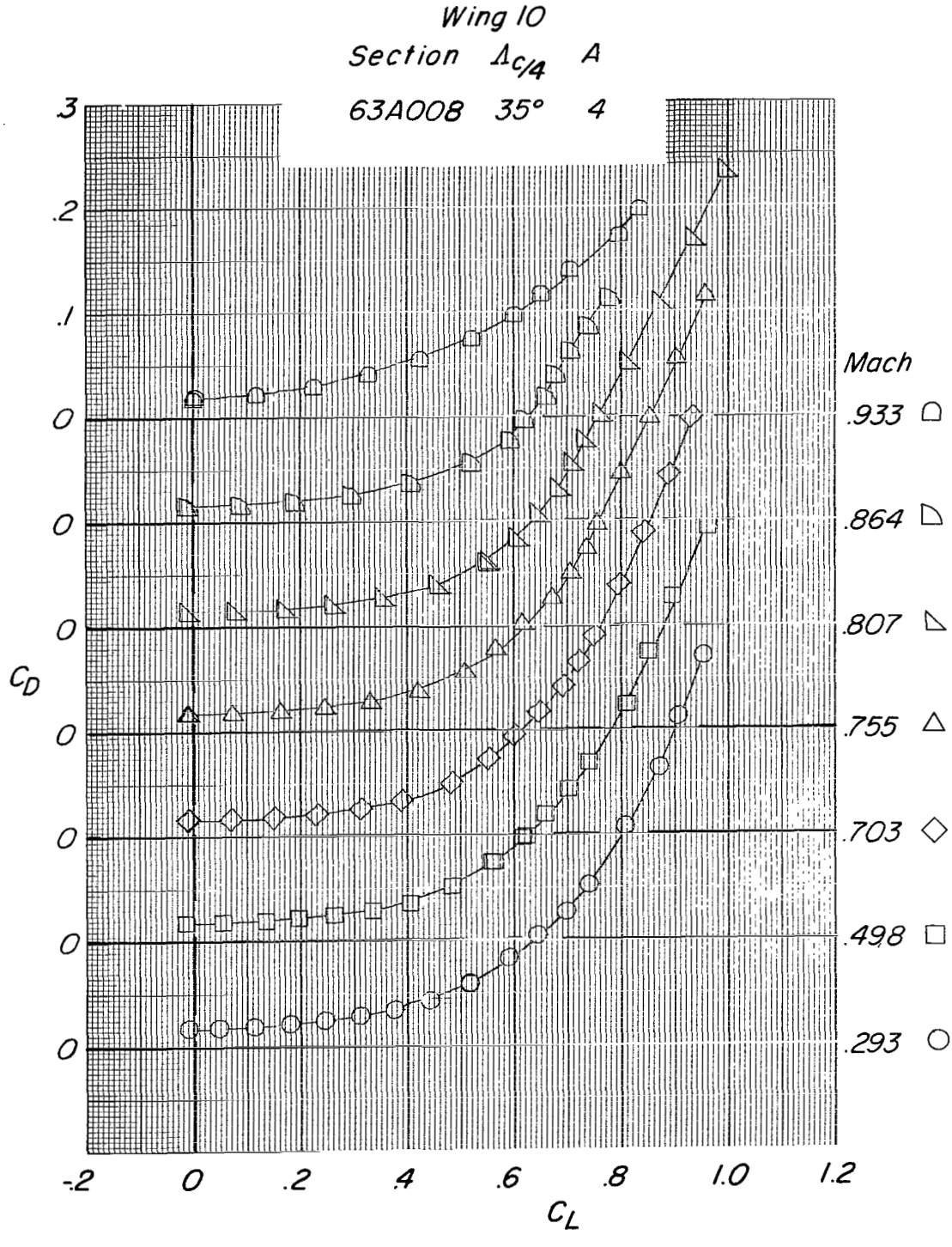
(c) α versus C_L.

Figure 16.- Continued.



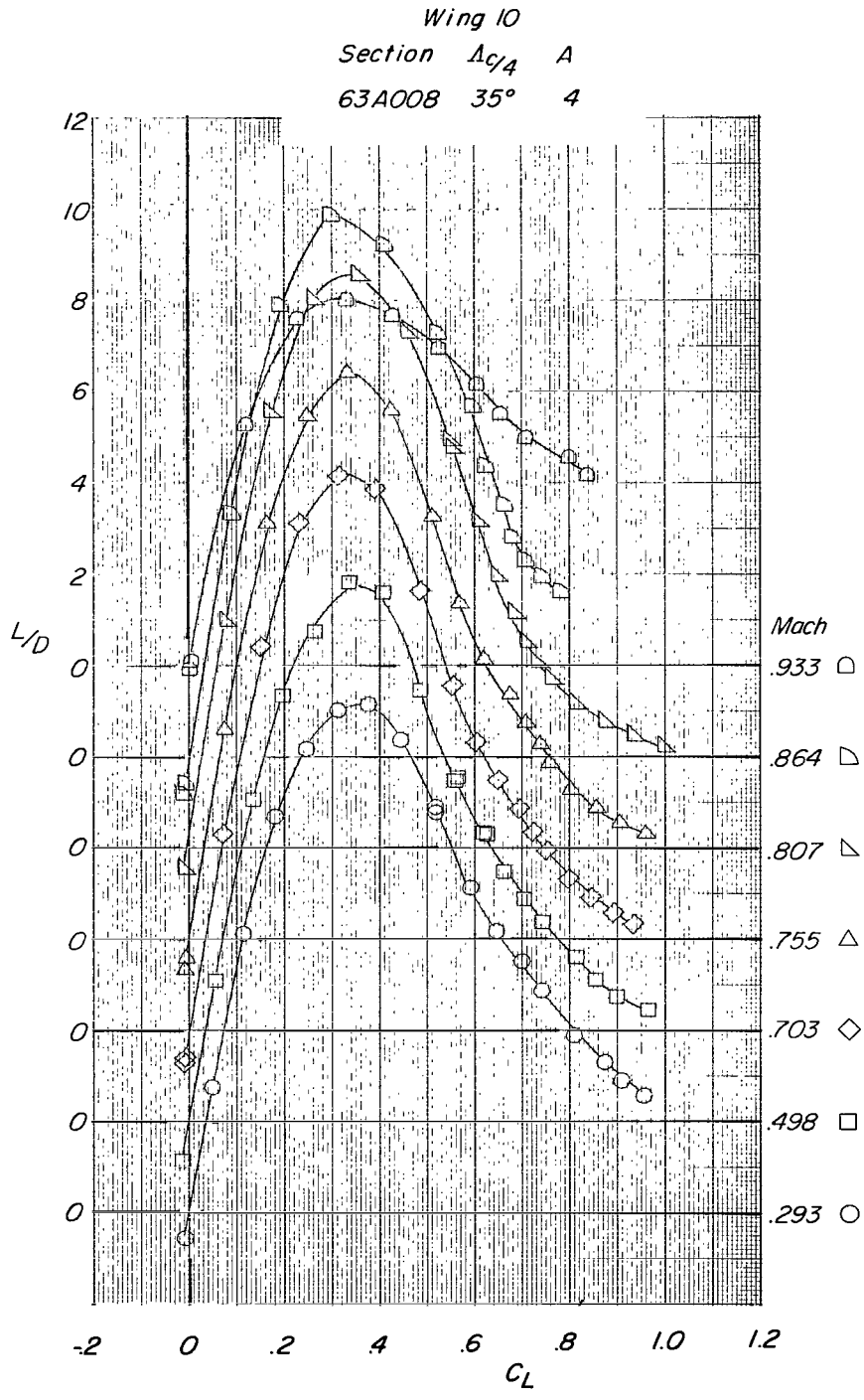
(d) C_m versus C_L.

Figure 16.- Continued.



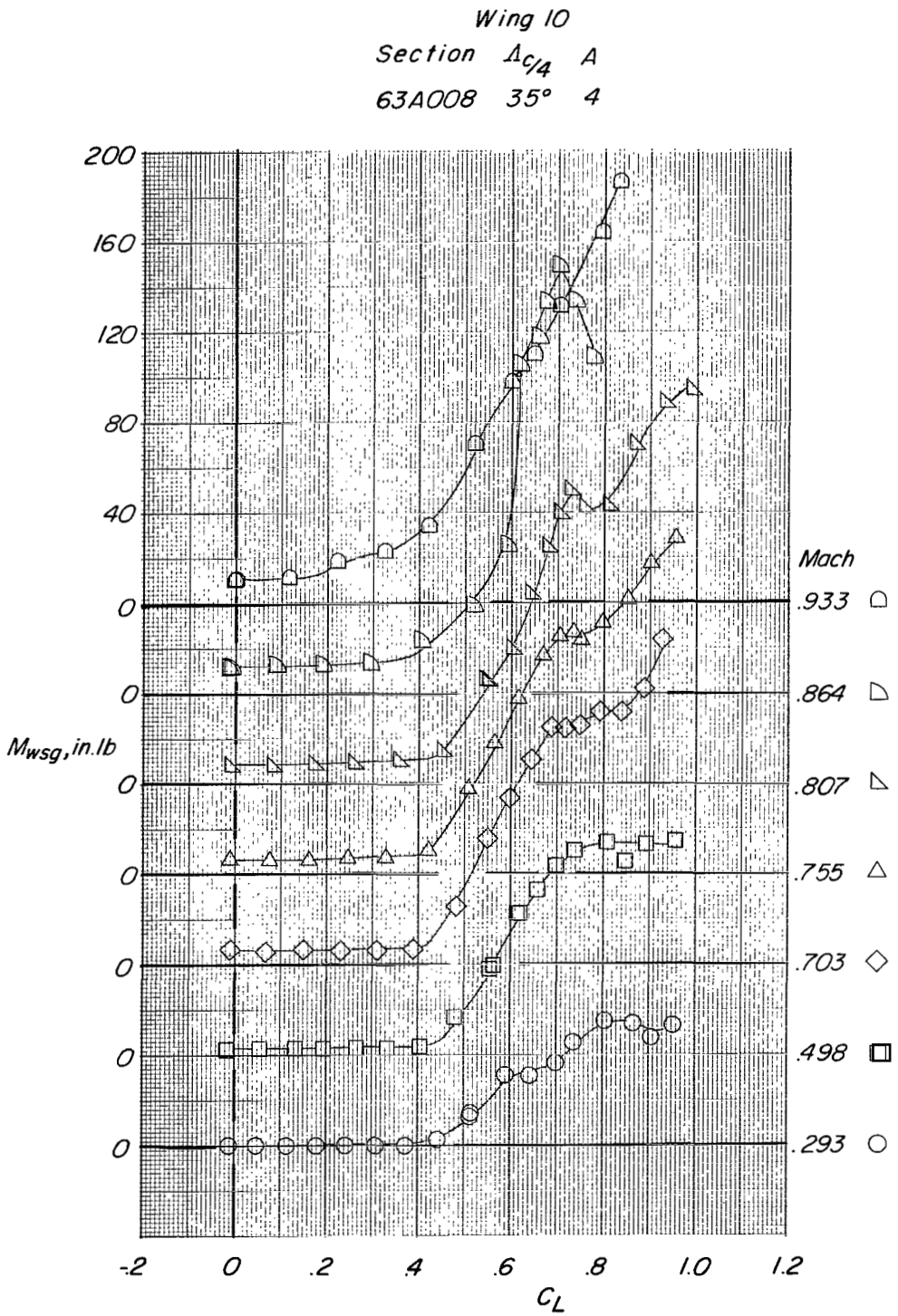
(e) C_D versus C_L.

Figure 16.- Continued.



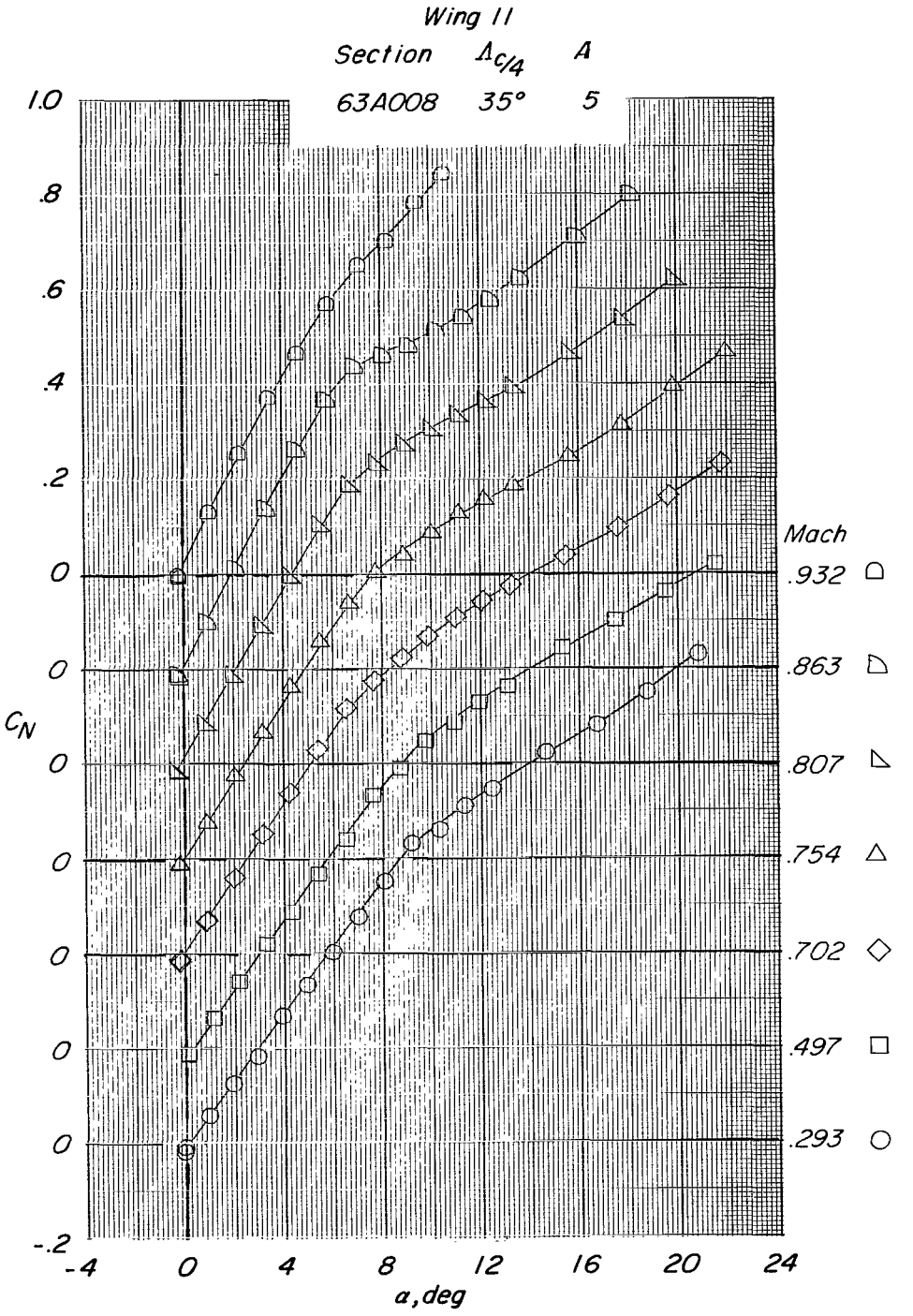
(f) L/D versus C_L.

Figure 16.- Continued.



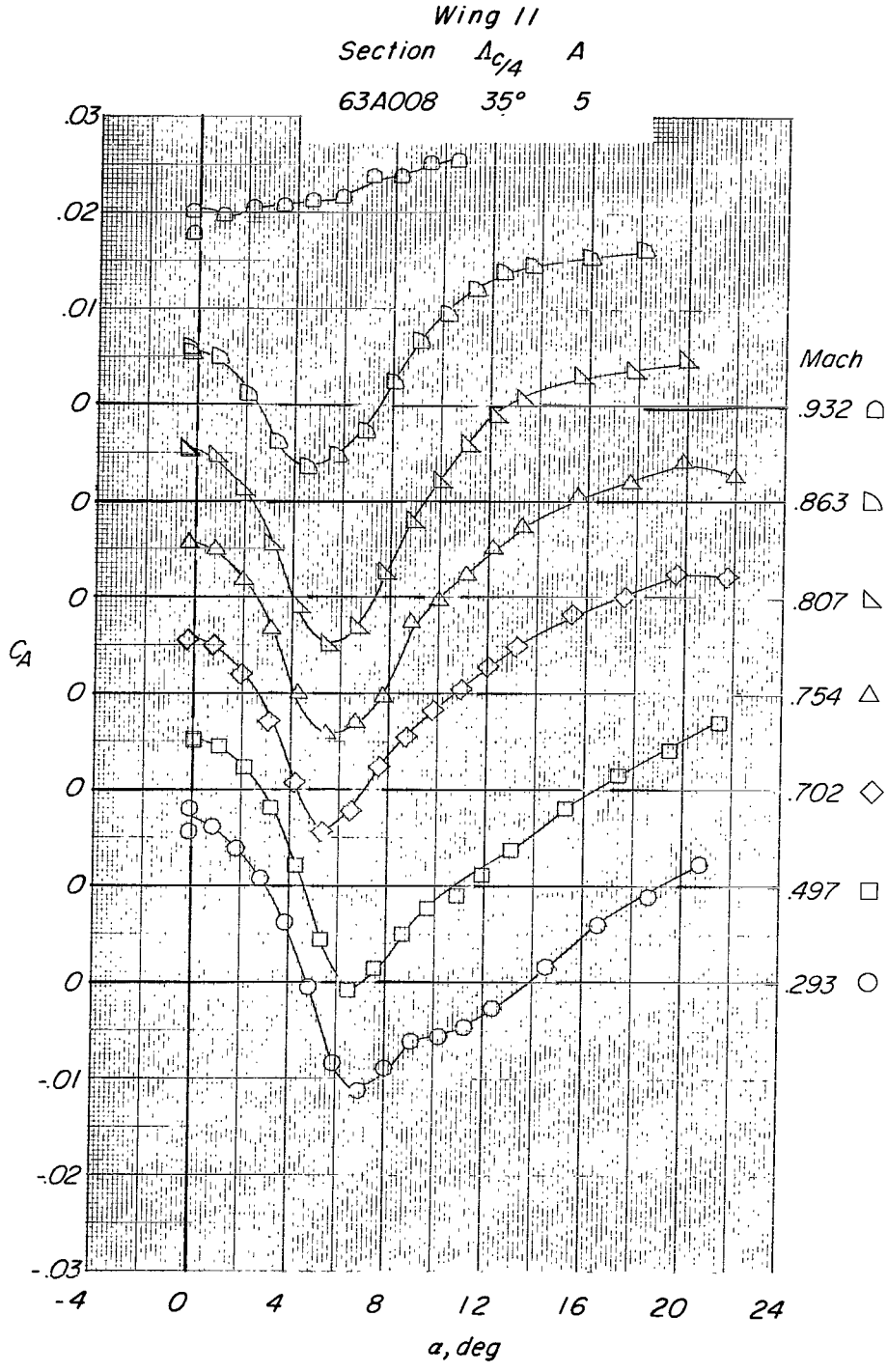
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 16.- Concluded.



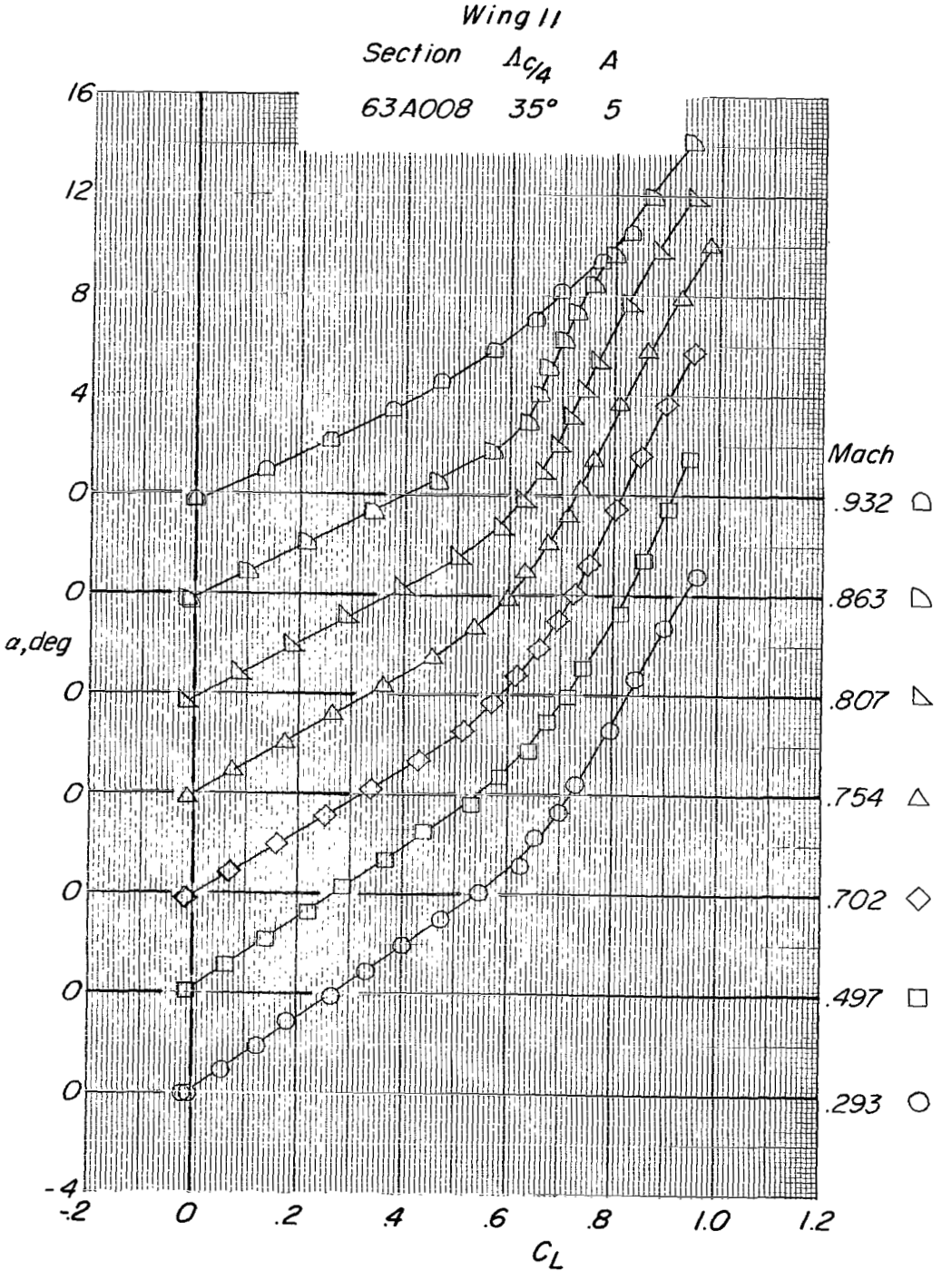
(a) C_N versus α.

Figure 17.- Static longitudinal aerodynamic and buffet characteristics of the wing 11 configuration at Mach numbers from 0.29 to 0.93. Rounded forebody; transition grit on.



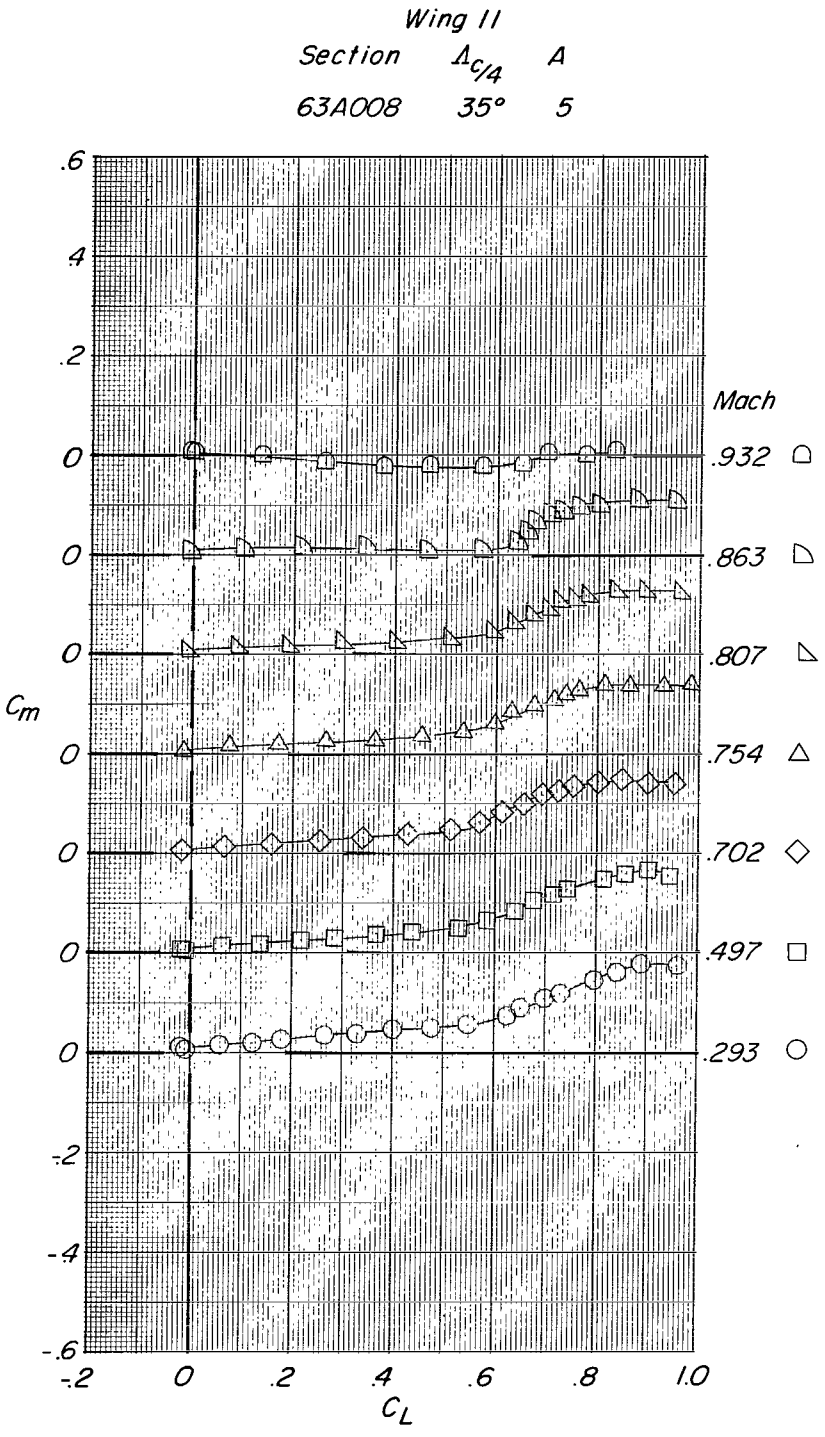
(b) C_A versus α.

Figure 17.- Continued.



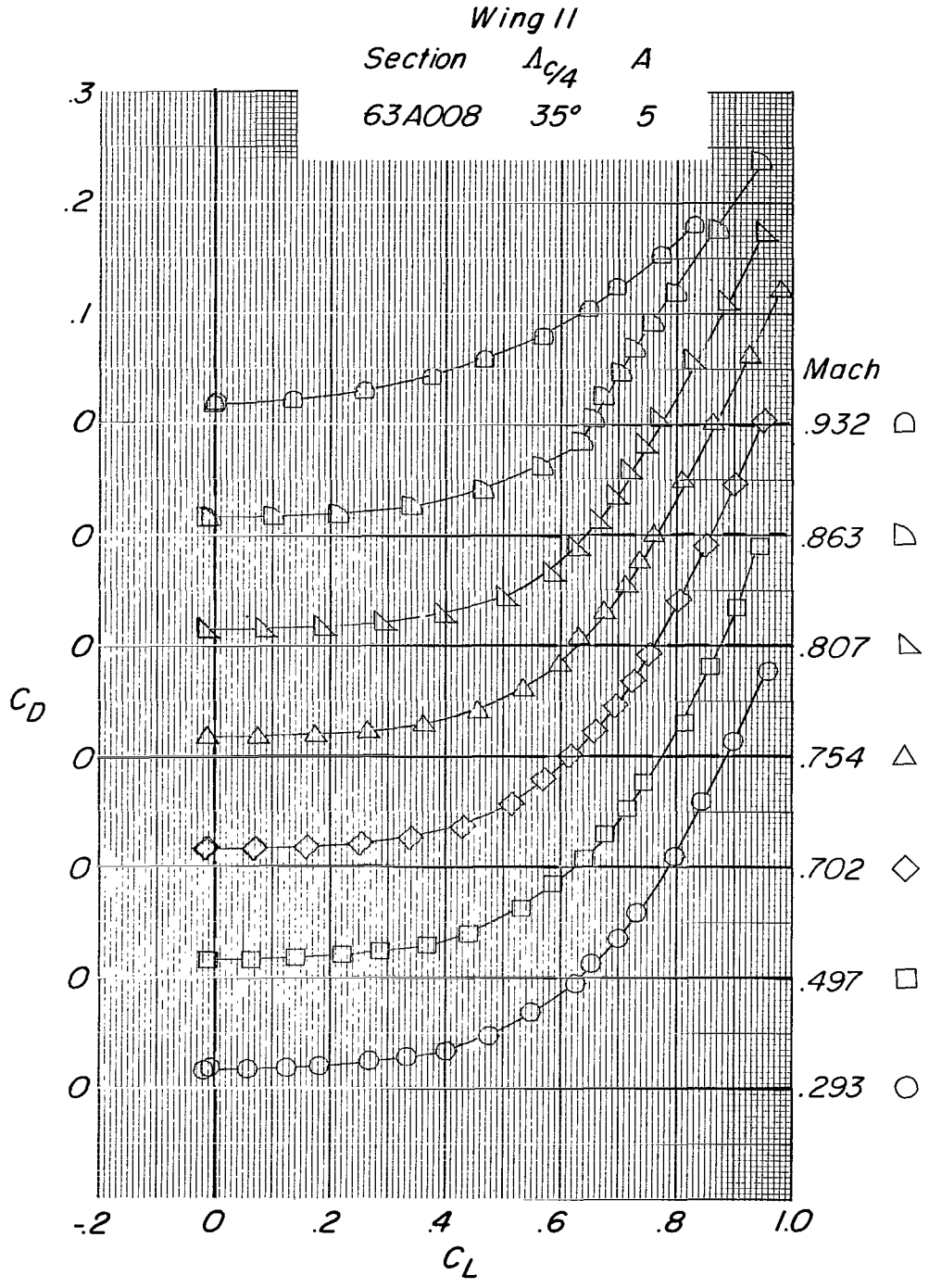
(c) α versus C_L.

Figure 17.- Continued.



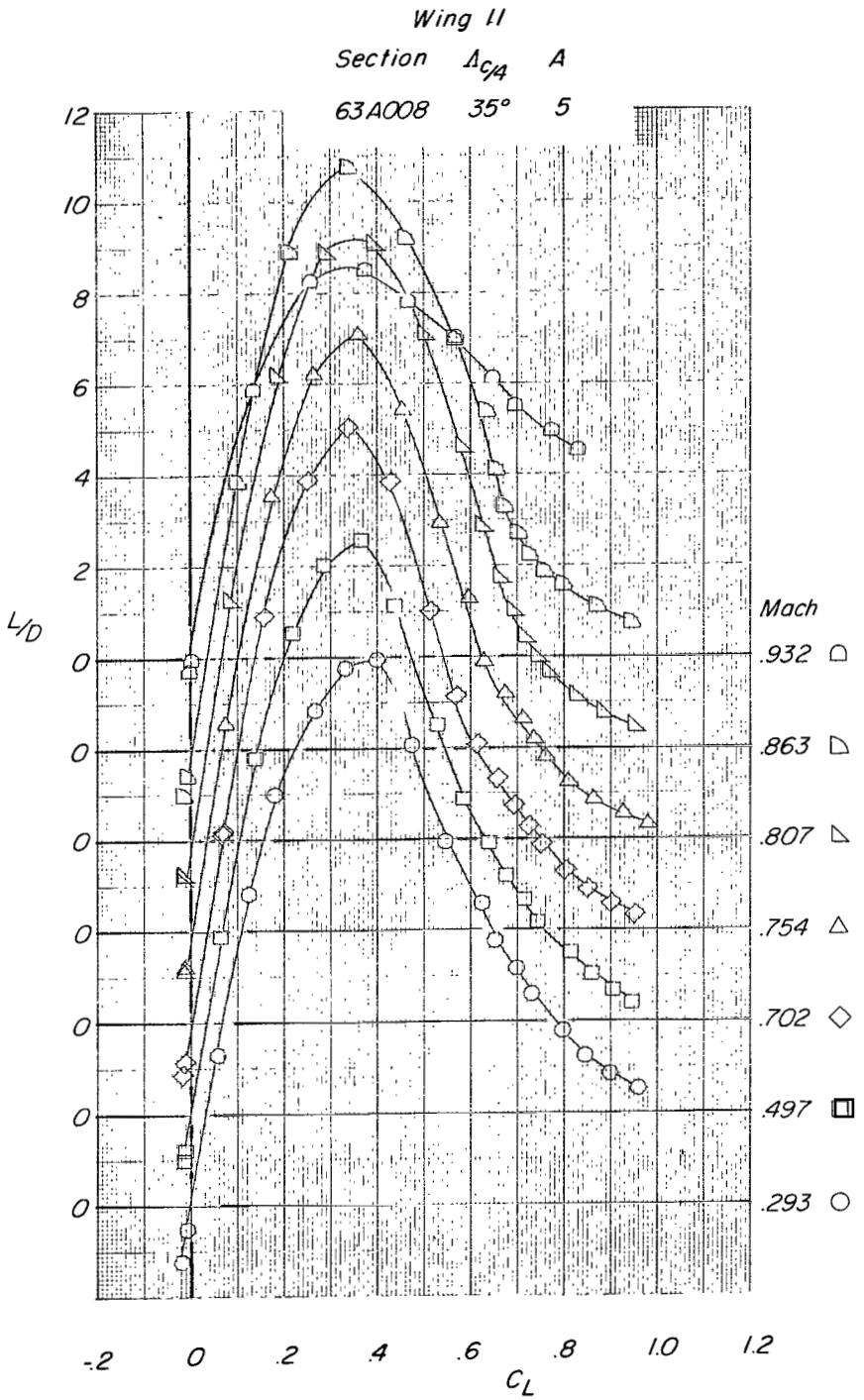
(d) C_m versus C_L.

Figure 17.- Continued.



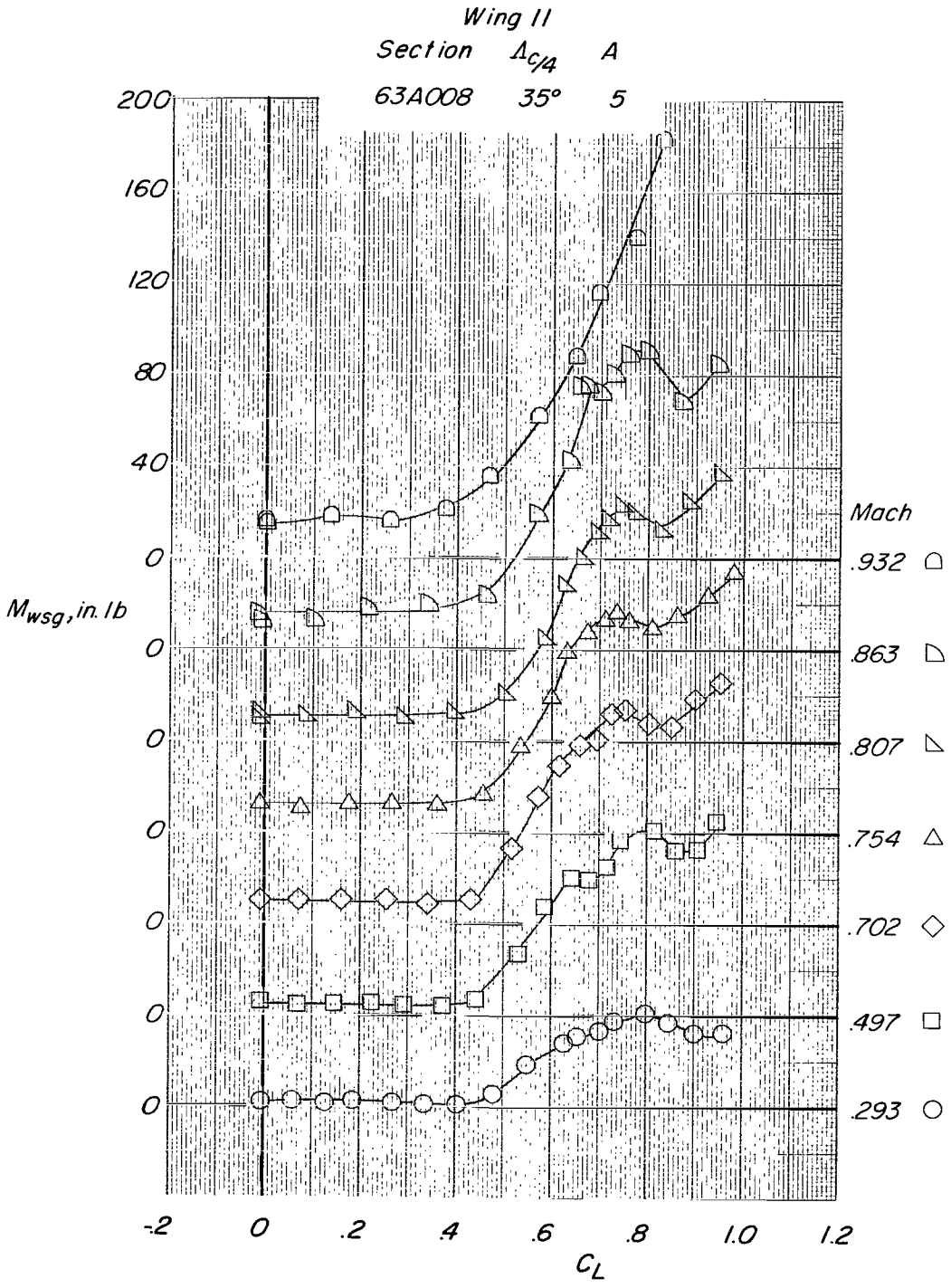
(e) C_D versus C_L.

Figure 17.- Continued.



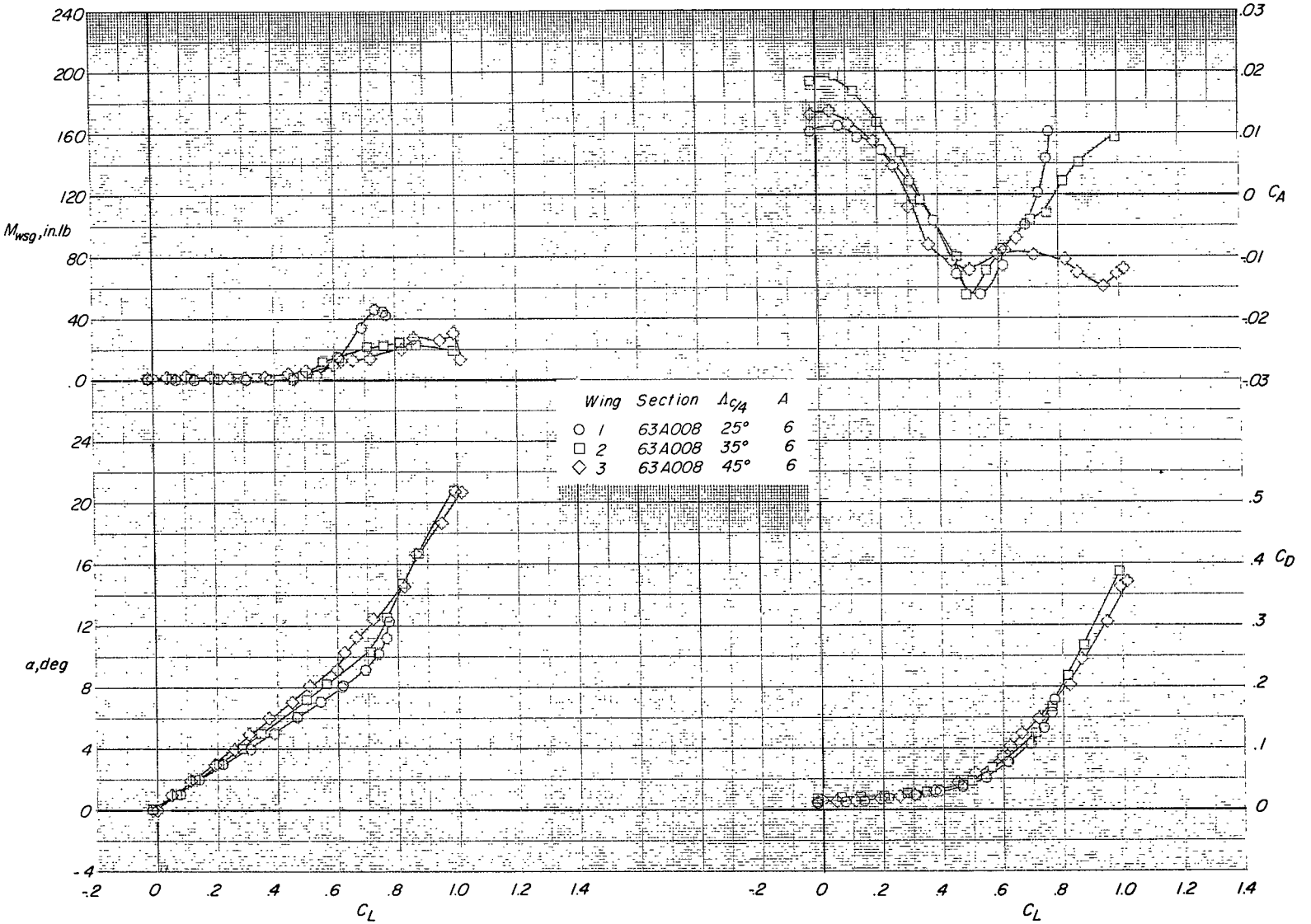
(f) L/D versus C_L.

Figure 17.- Continued.



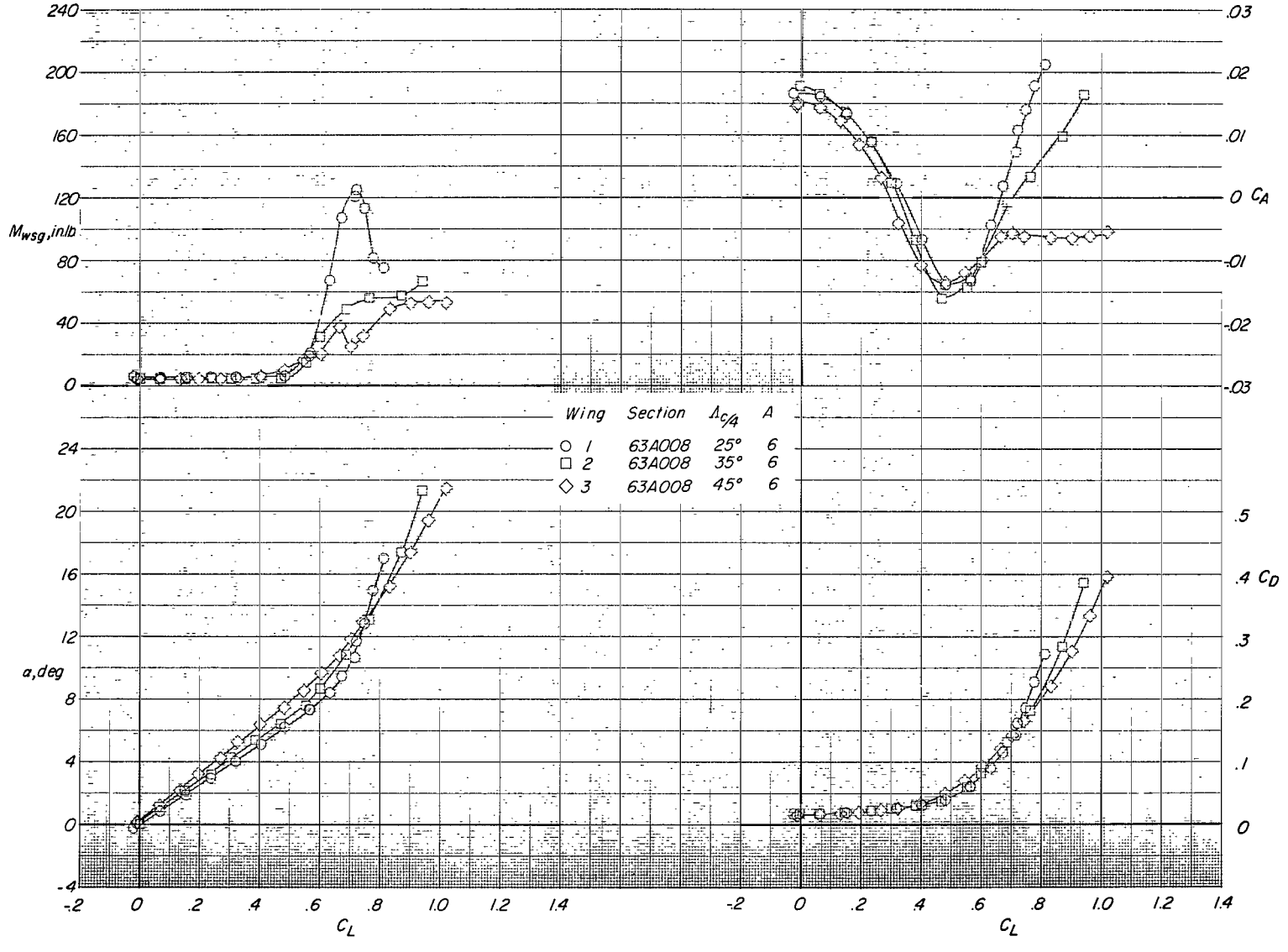
(g) M_{wsg} versus C_L. (1 in. lb = 0.113 m-N.)

Figure 17.- Concluded.



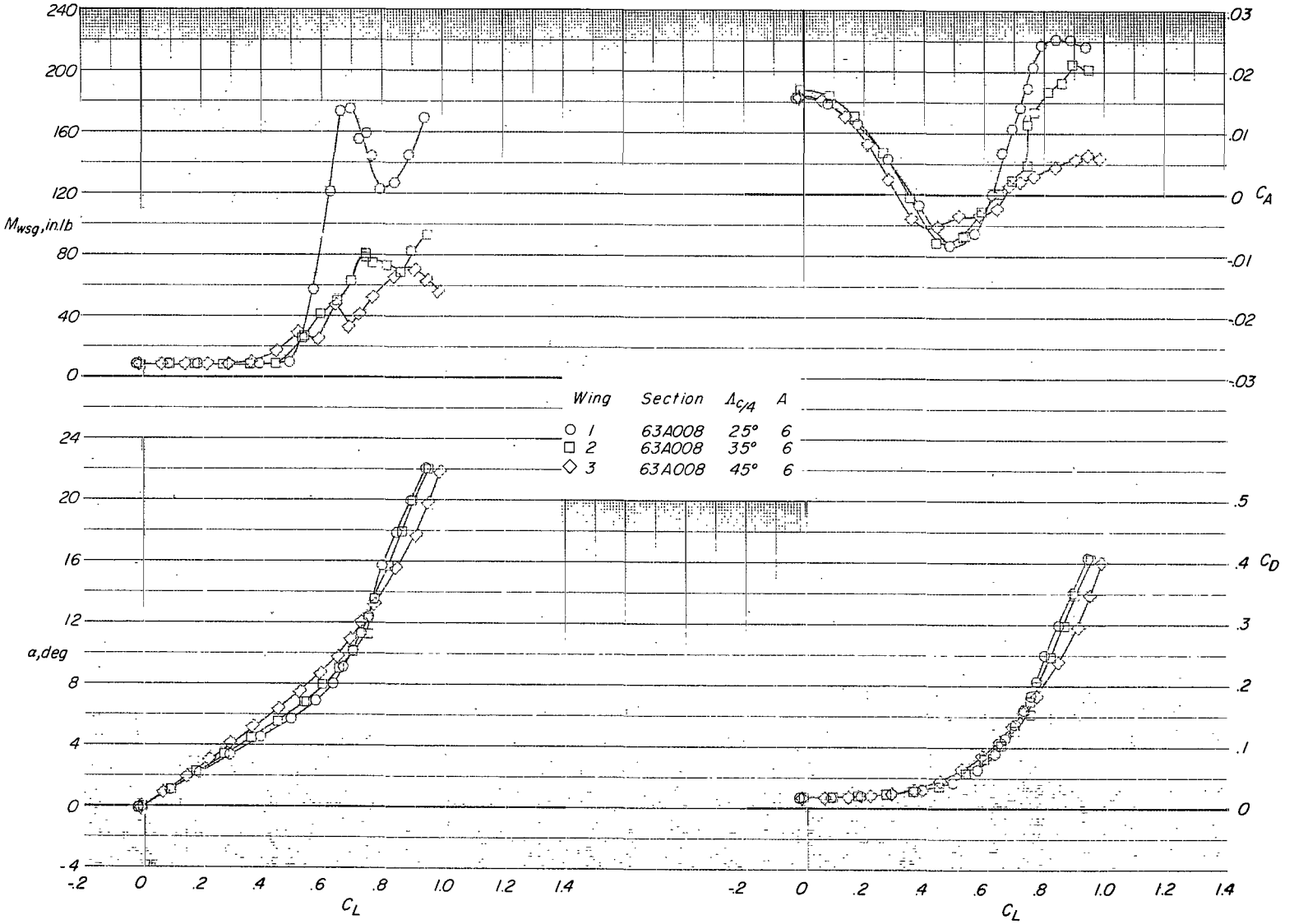
(a) Mach number, 0.23.

Figure 18.- Effect of wing sweep on the variations of M_{wsg}, α, C_A, and C_D with C_L. (1 in. lb = 0.113 m-N.)



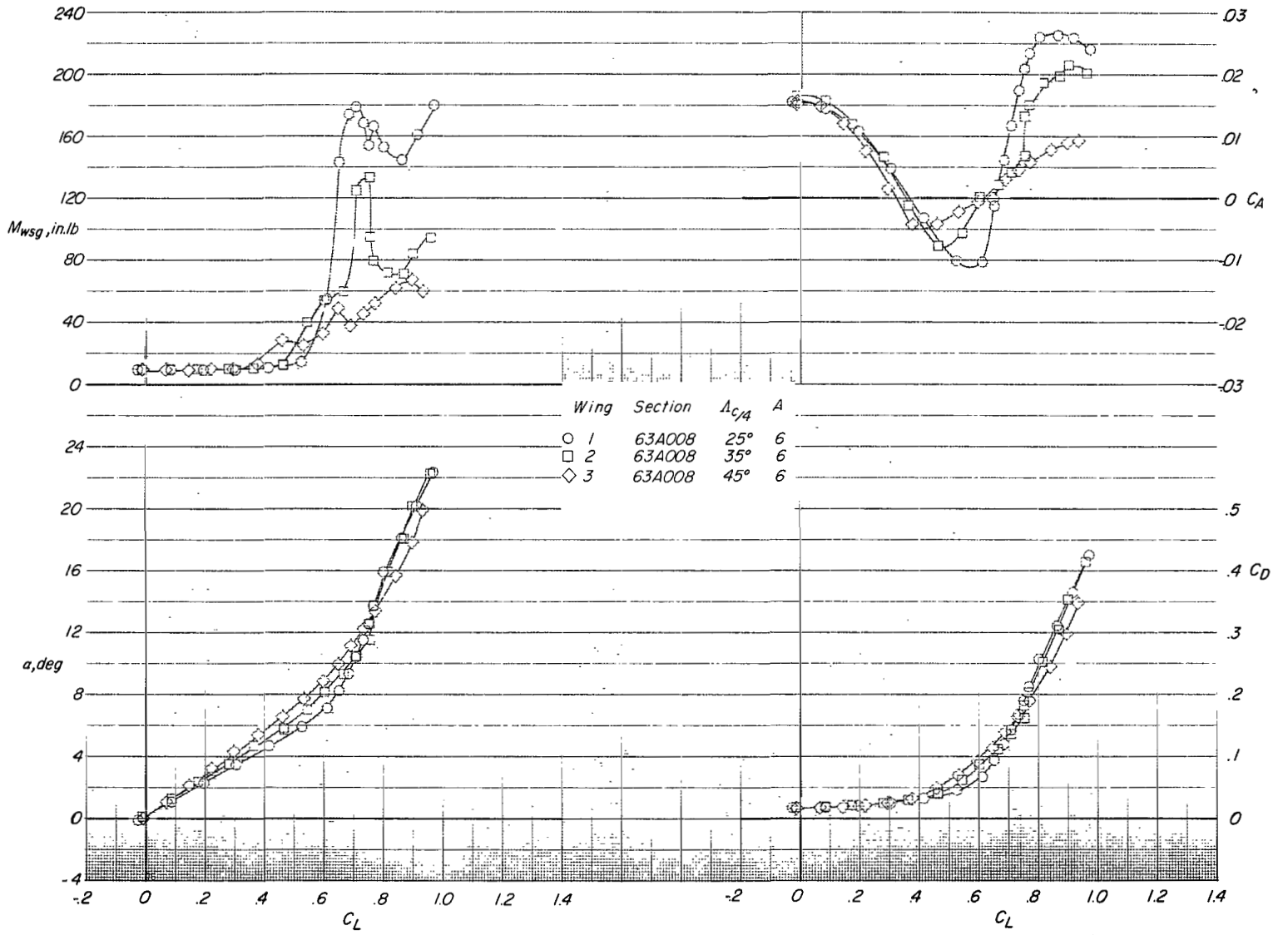
(b) Mach number, 0.46.

Figure 18.- Continued.



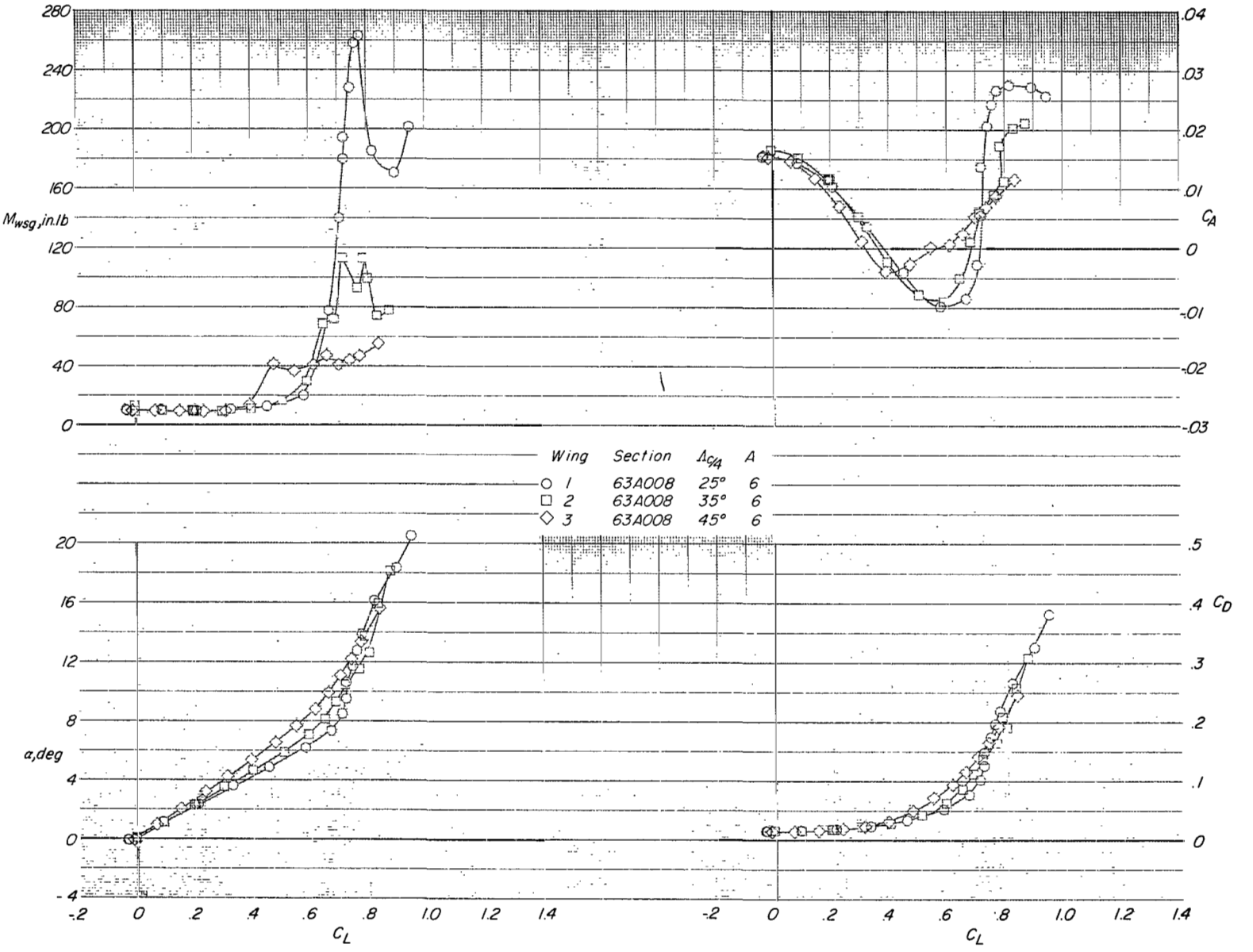
(c) Mach number, 0.67.

Figure 18.- Continued.



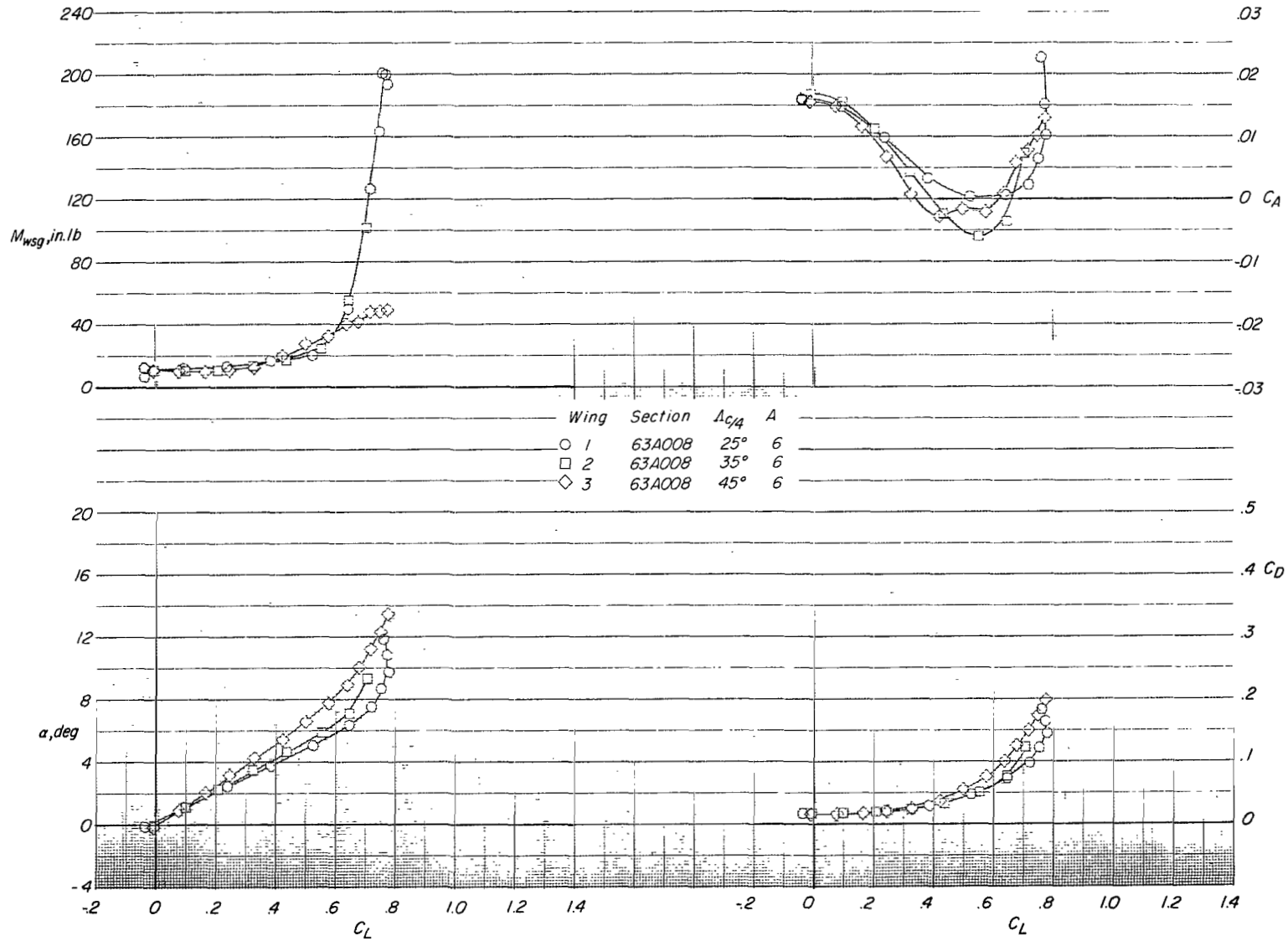
(d) Mach number, 0.72.

Figure 18.- Continued.



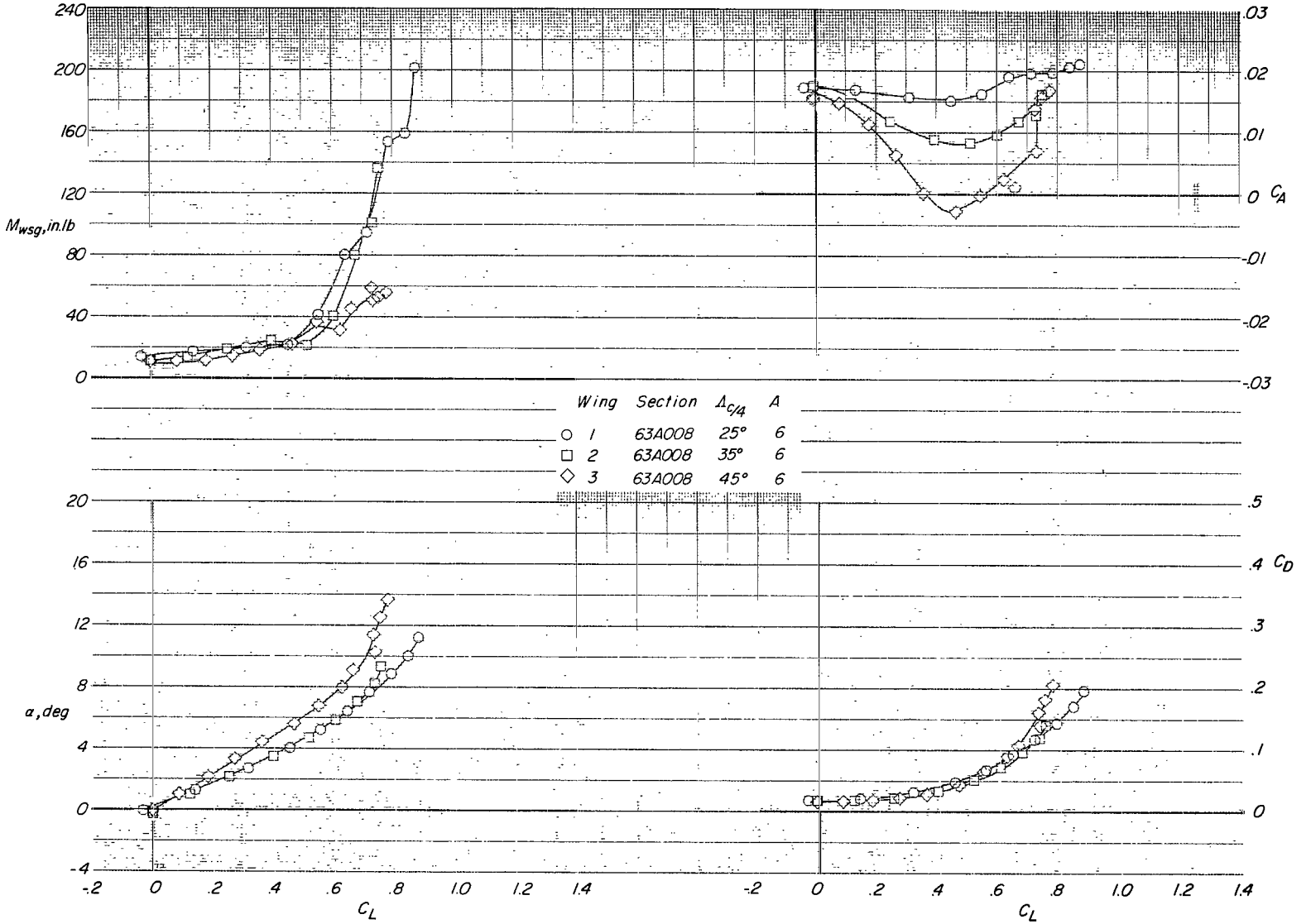
(e) Mach number, 0.77.

Figure 18.- Continued.



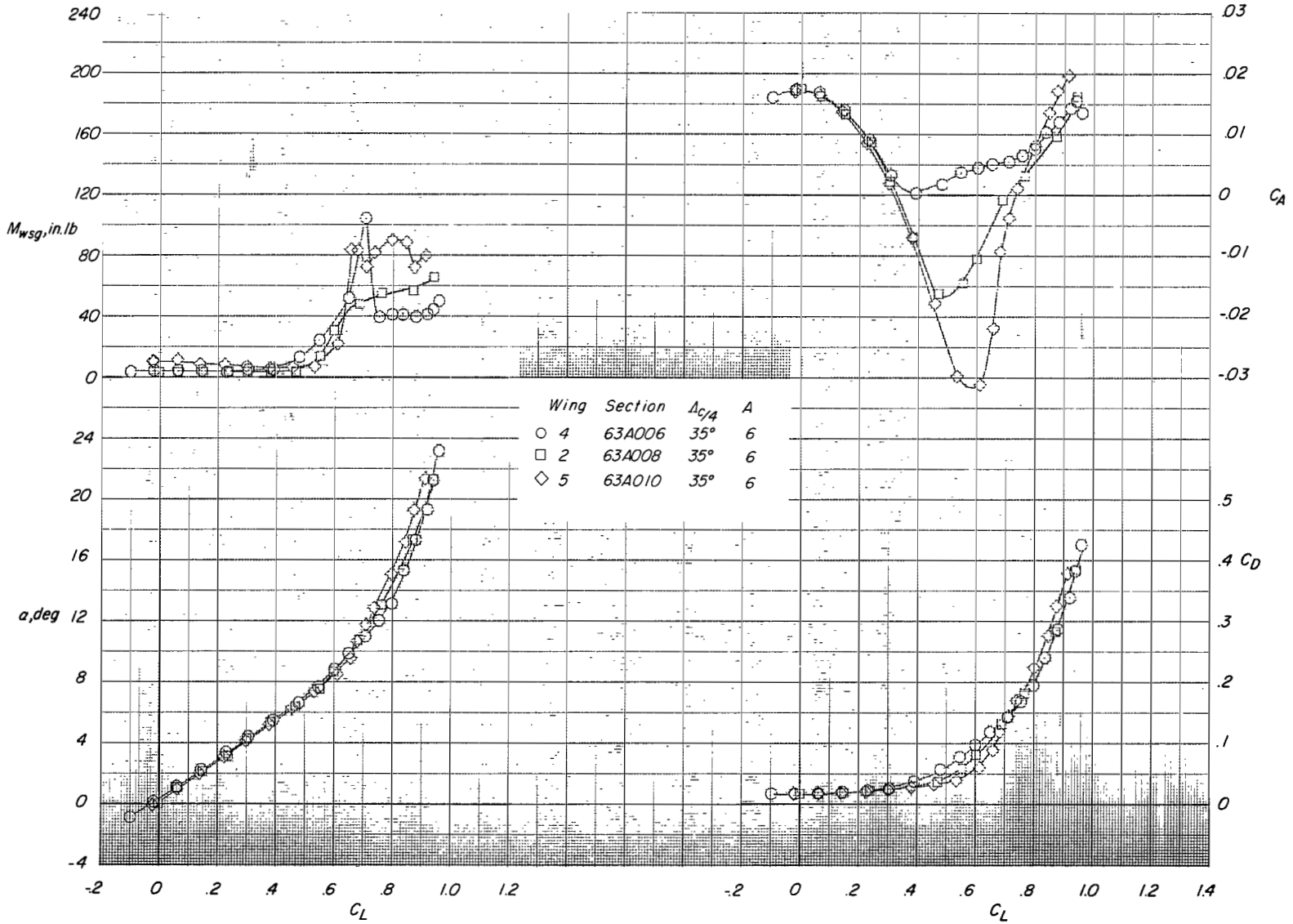
(f) Mach number, 0.82.

Figure 18.- Continued.



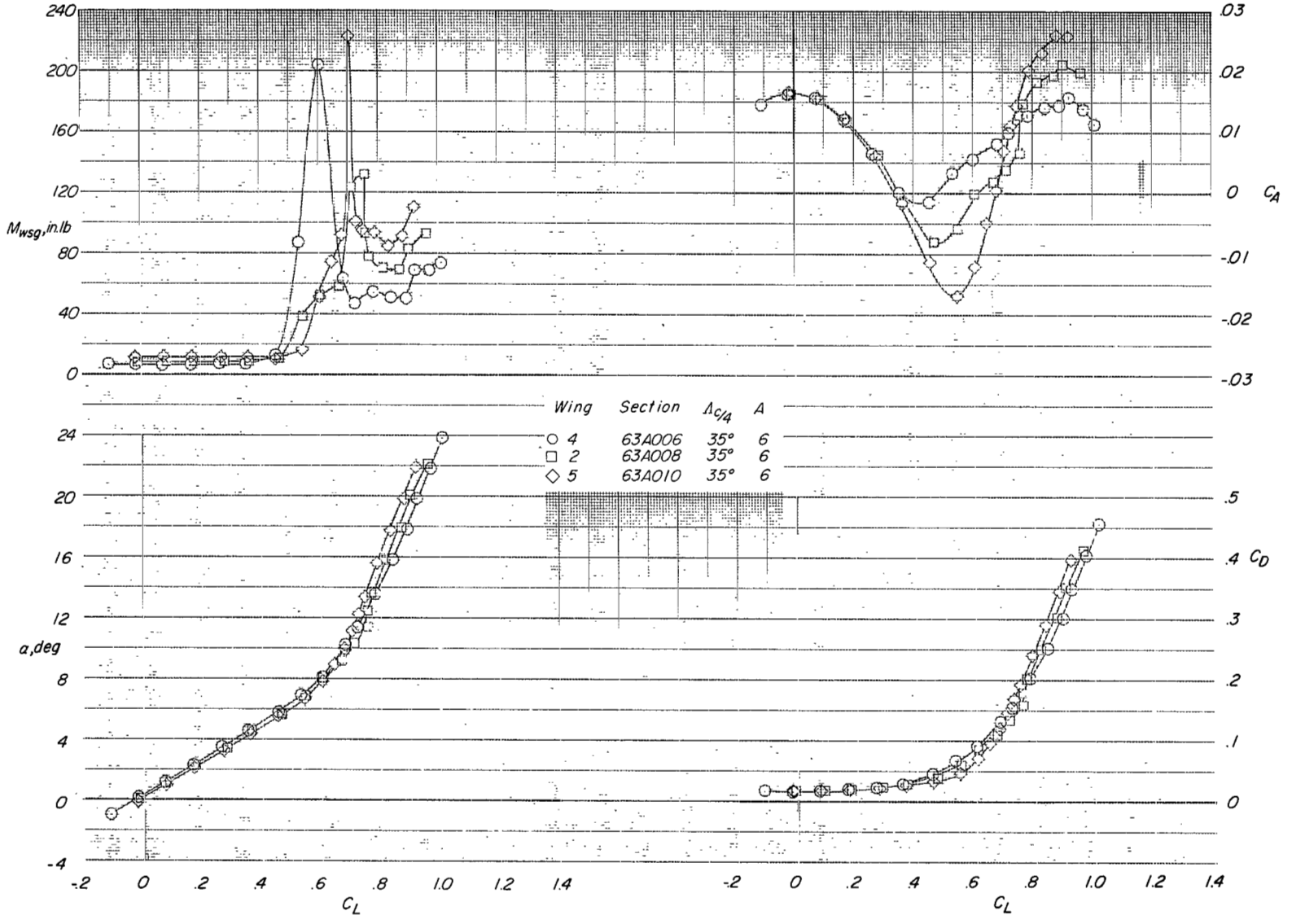
(g) Mach number, 0.88.

Figure 18.- Concluded.



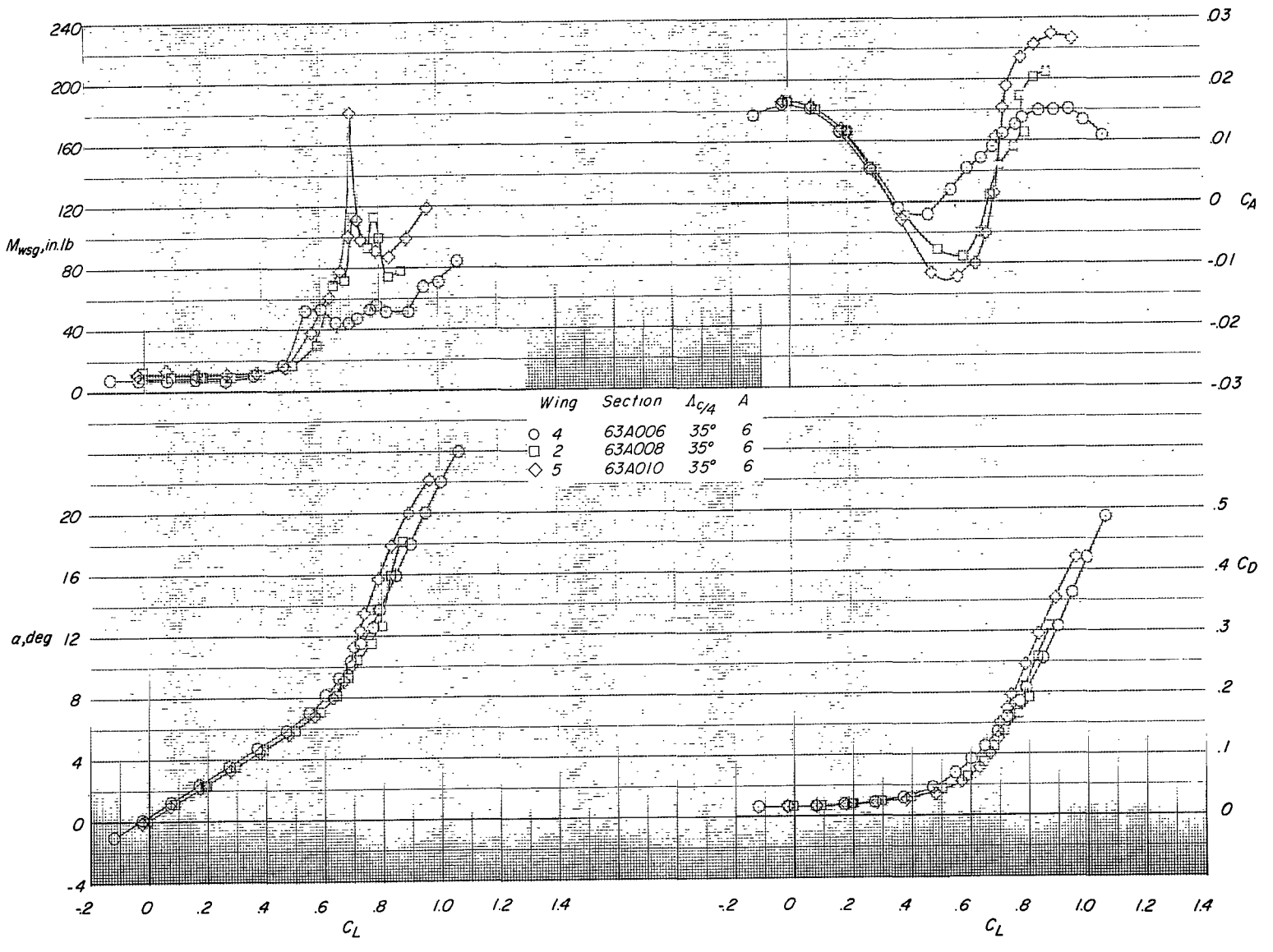
(a) Mach number, 0.46 for wing 2; 0.50 for wings 4 and 5.

Figure 19.- Effect of thickness-to-chord ratio on the variations of M_{wsg}, α, C_A, and C_D with C_L. (1 in. lb = 0.113 m-N.)



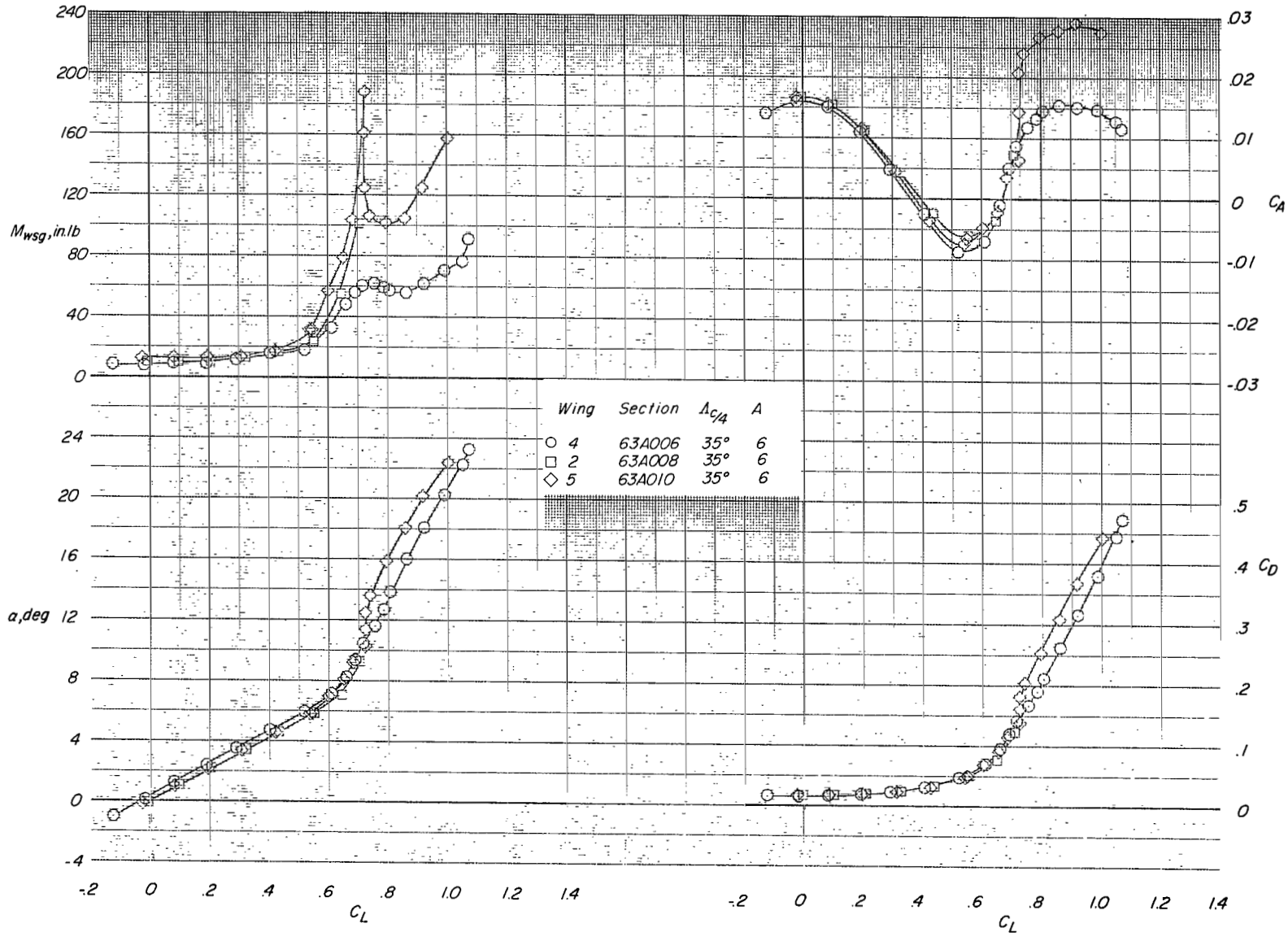
(b) Mach number, 0.71.

Figure 19.- Continued.



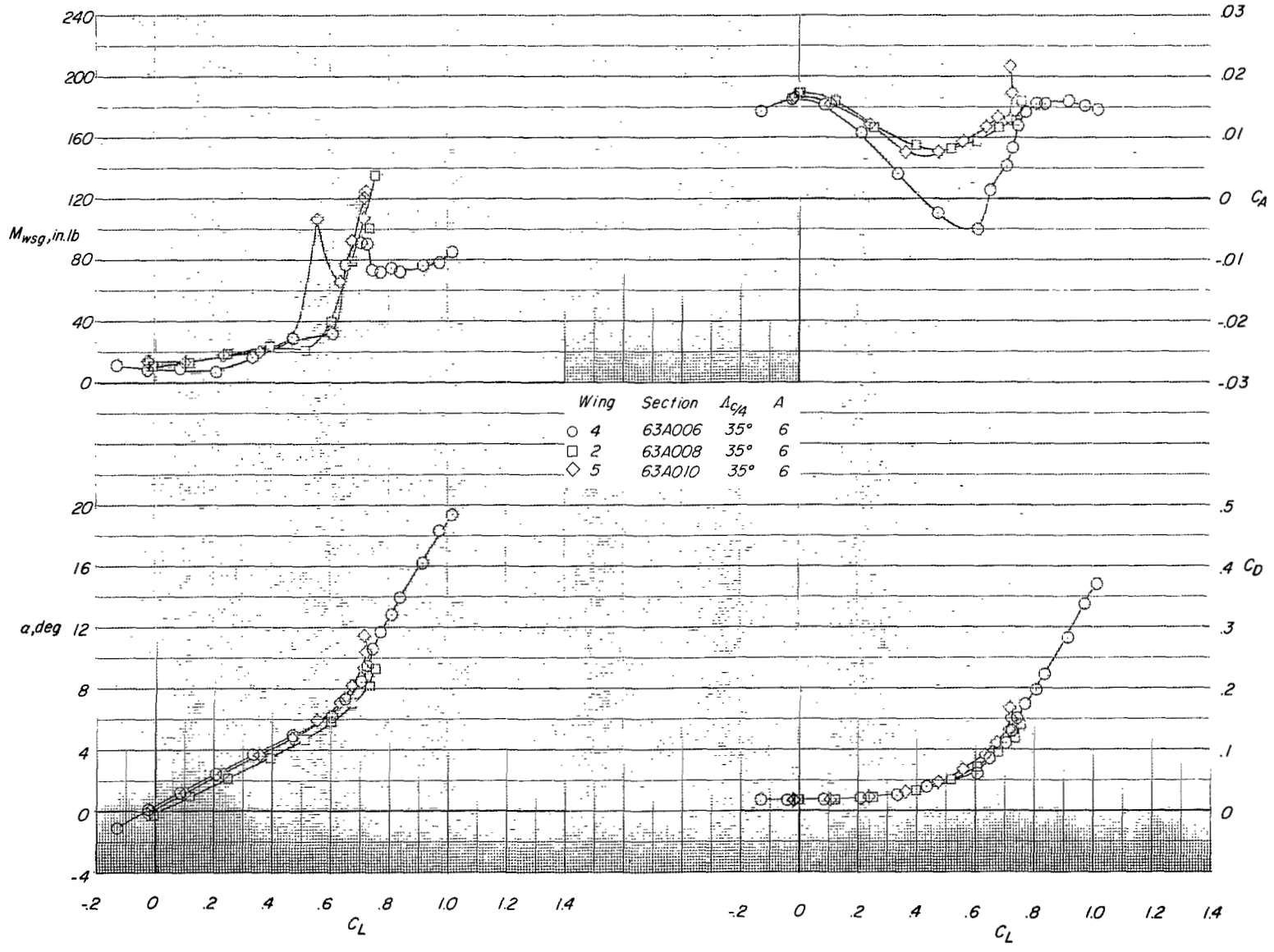
(c) Mach number, 0.76.

Figure 19.- Continued.



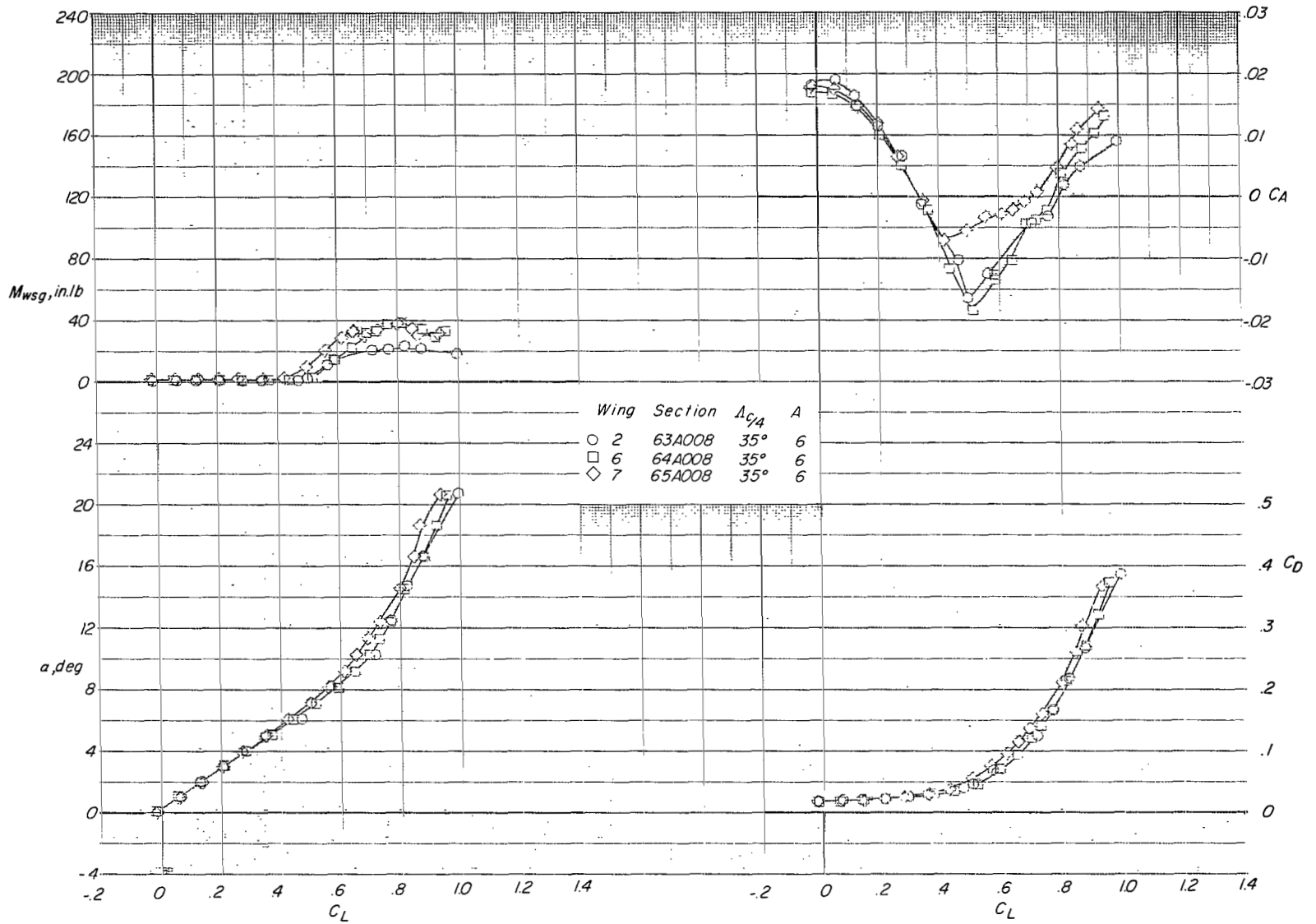
(d) Mach number, 0.81.

Figure 19.- Continued.



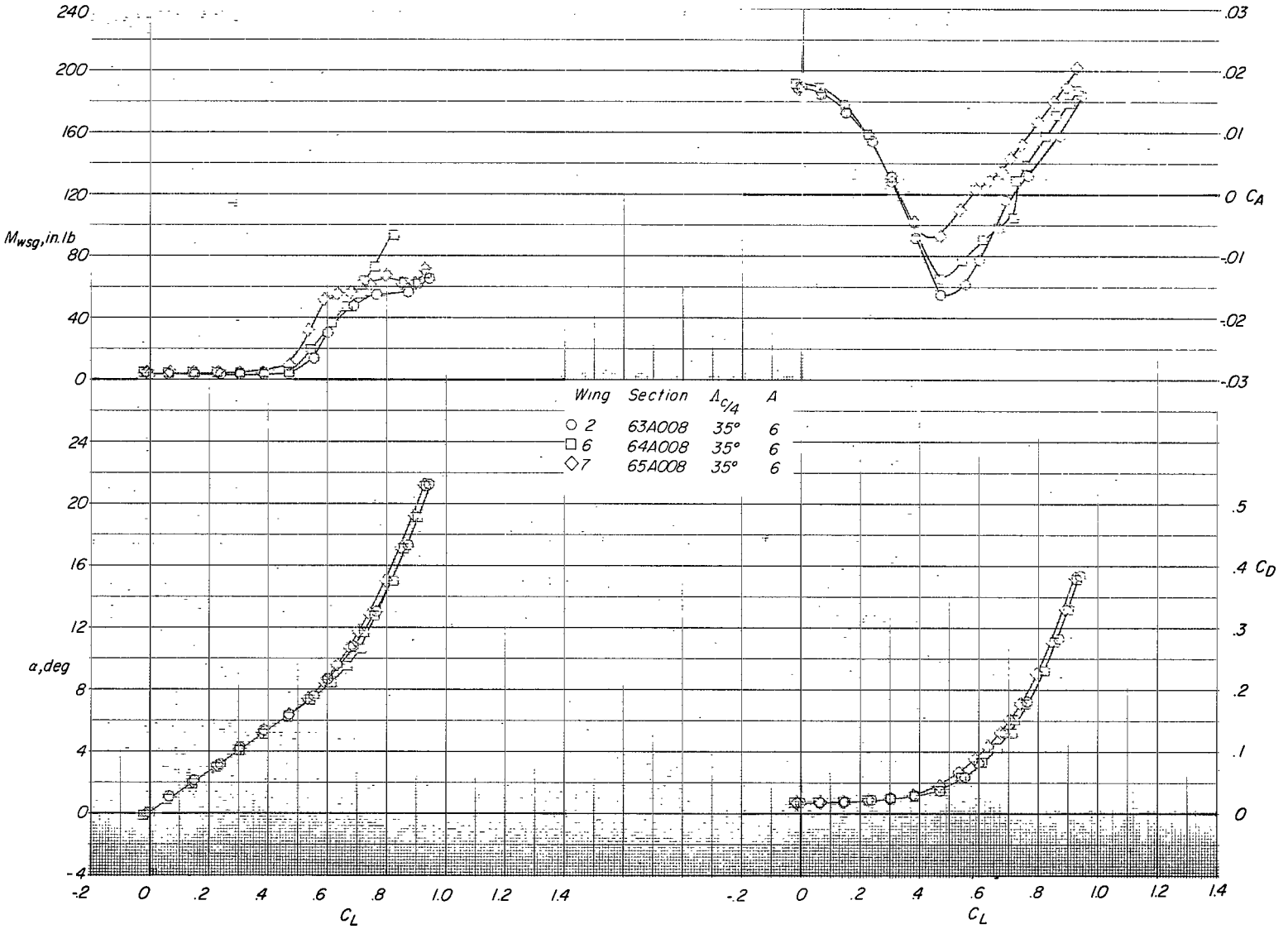
(e) Mach number, 0.88 for wing 2; 0.87 for wing 4; and 0.86 for wing 5.

Figure 19.- Concluded.



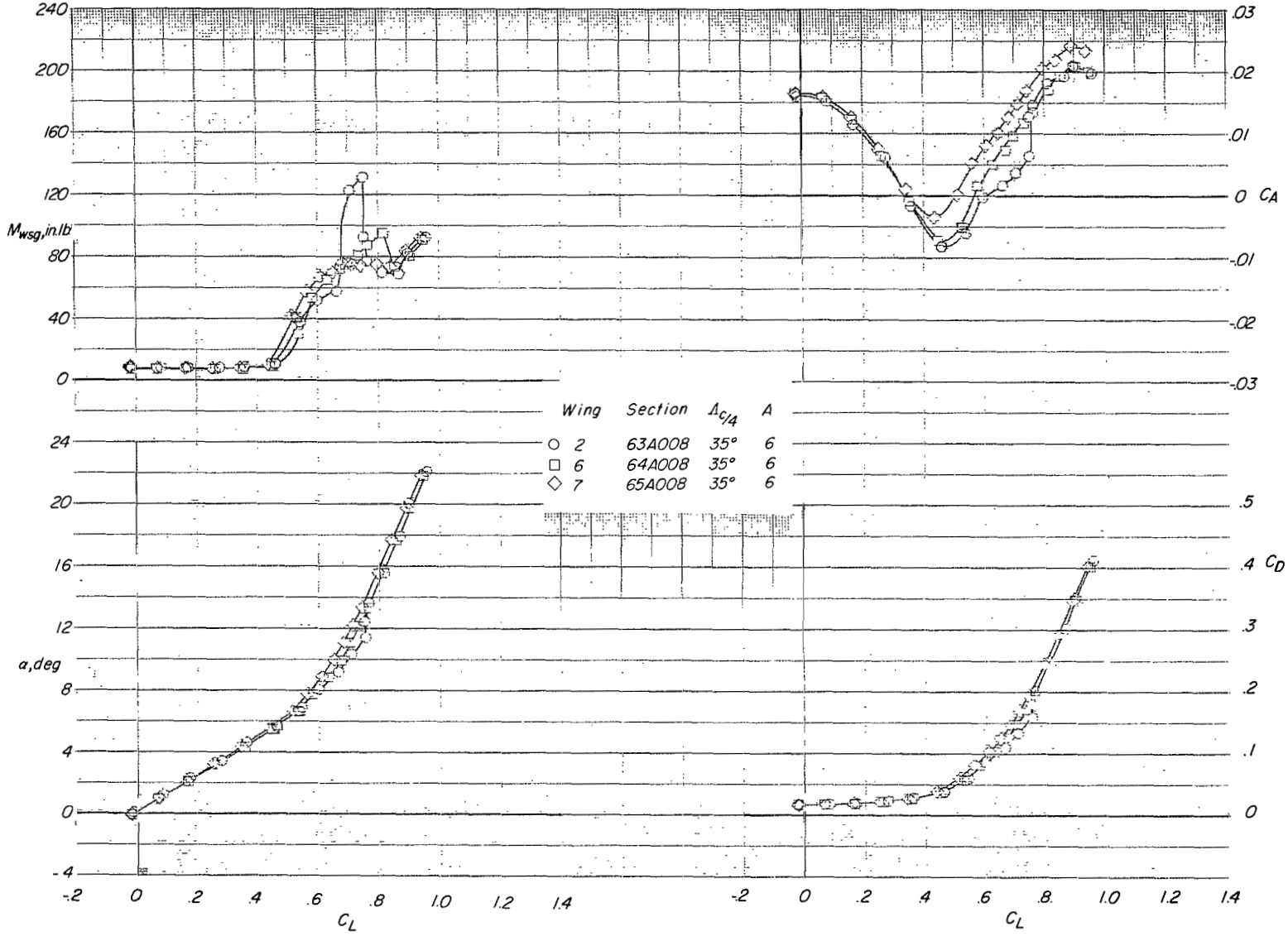
(a) Mach number, 0.23 for wing 2; 0.29 for wings 6 and 7.

Figure 20.- Effect of position of maximum thickness on the variations of M_{wsg}, α, C_A, and C_D with C_L. (1 in. lb = 0.113 m-N.)



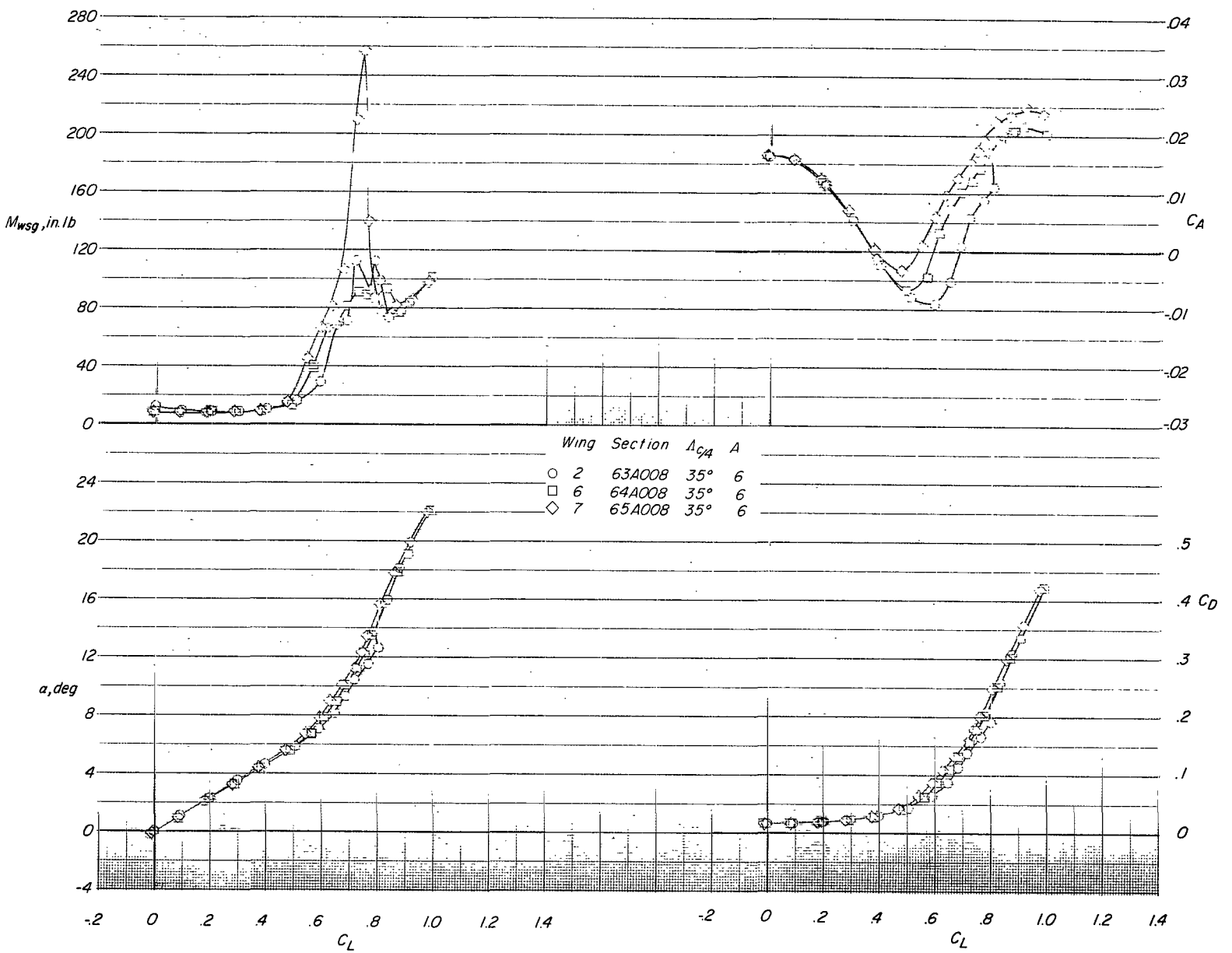
(b) Mach number, 0.46 for wing 2; 0.50 for wings 6 and 7.

Figure 20.- Continued.



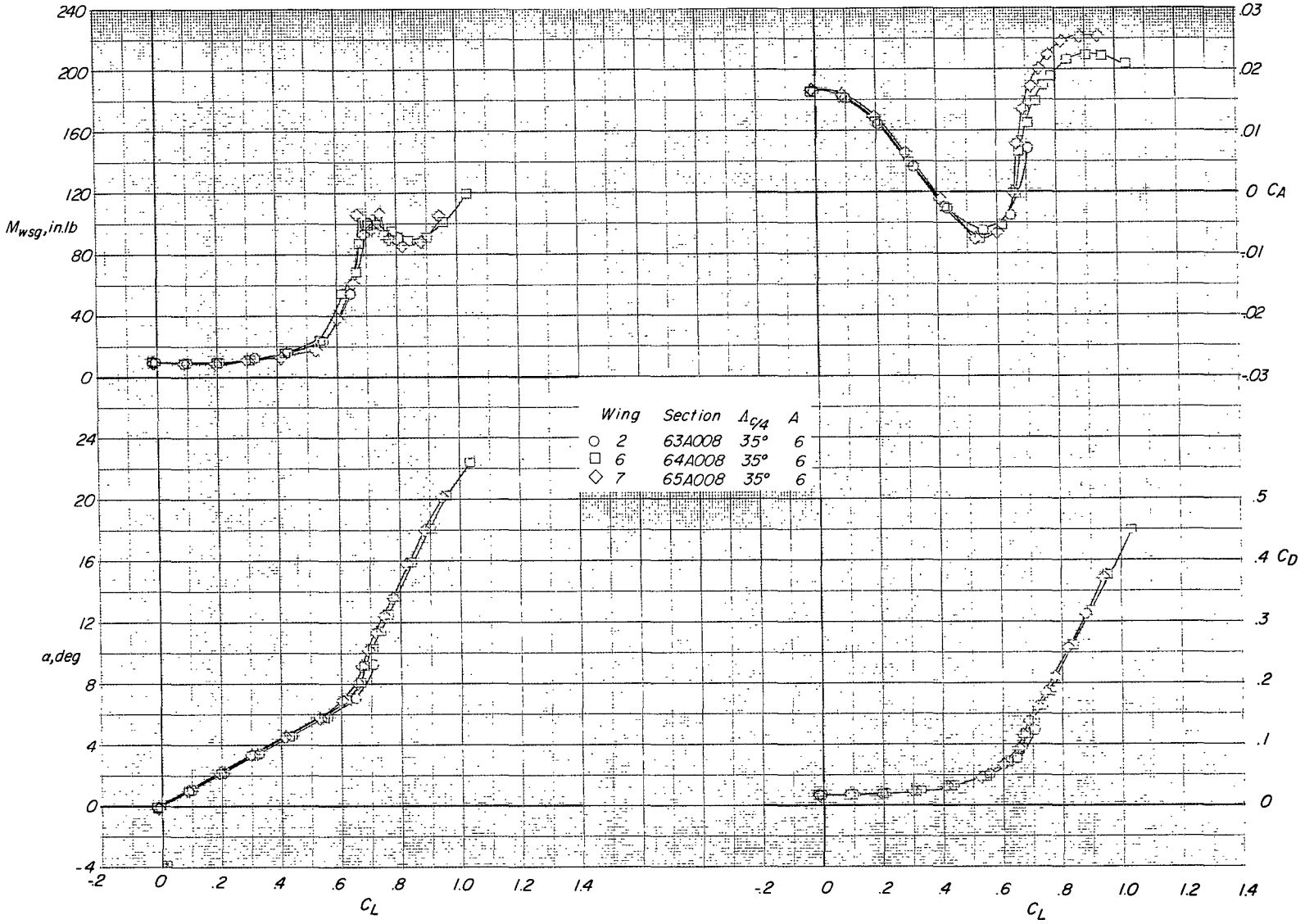
(c) Mach number, 0.72 for wing 2; 0.70 for wings 6 and 7.

Figure 20.- Continued.



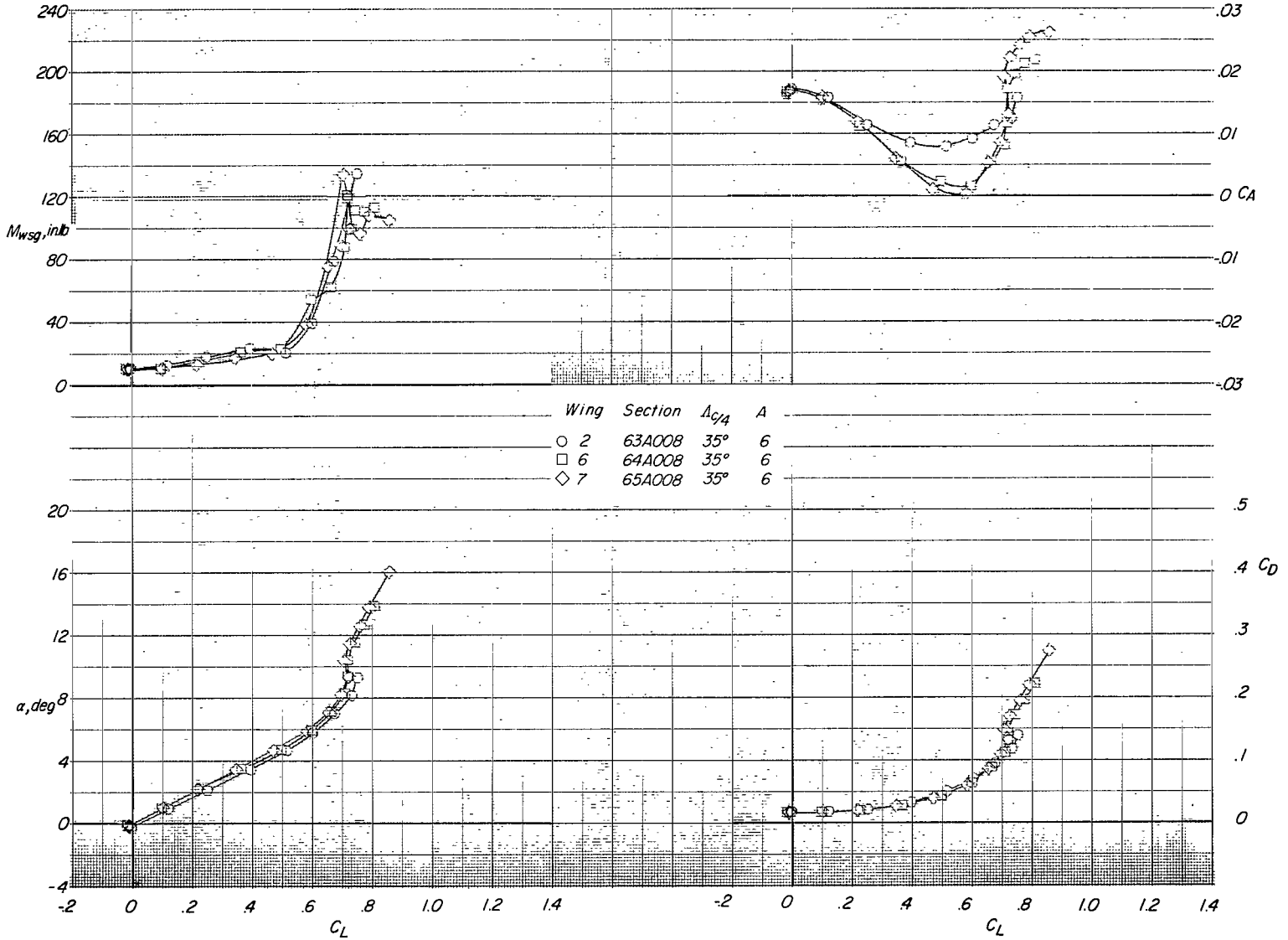
(d) Mach number, 0.77 for wing 2; 0.76 for wings 6 and 7.

Figure 20.- Continued.



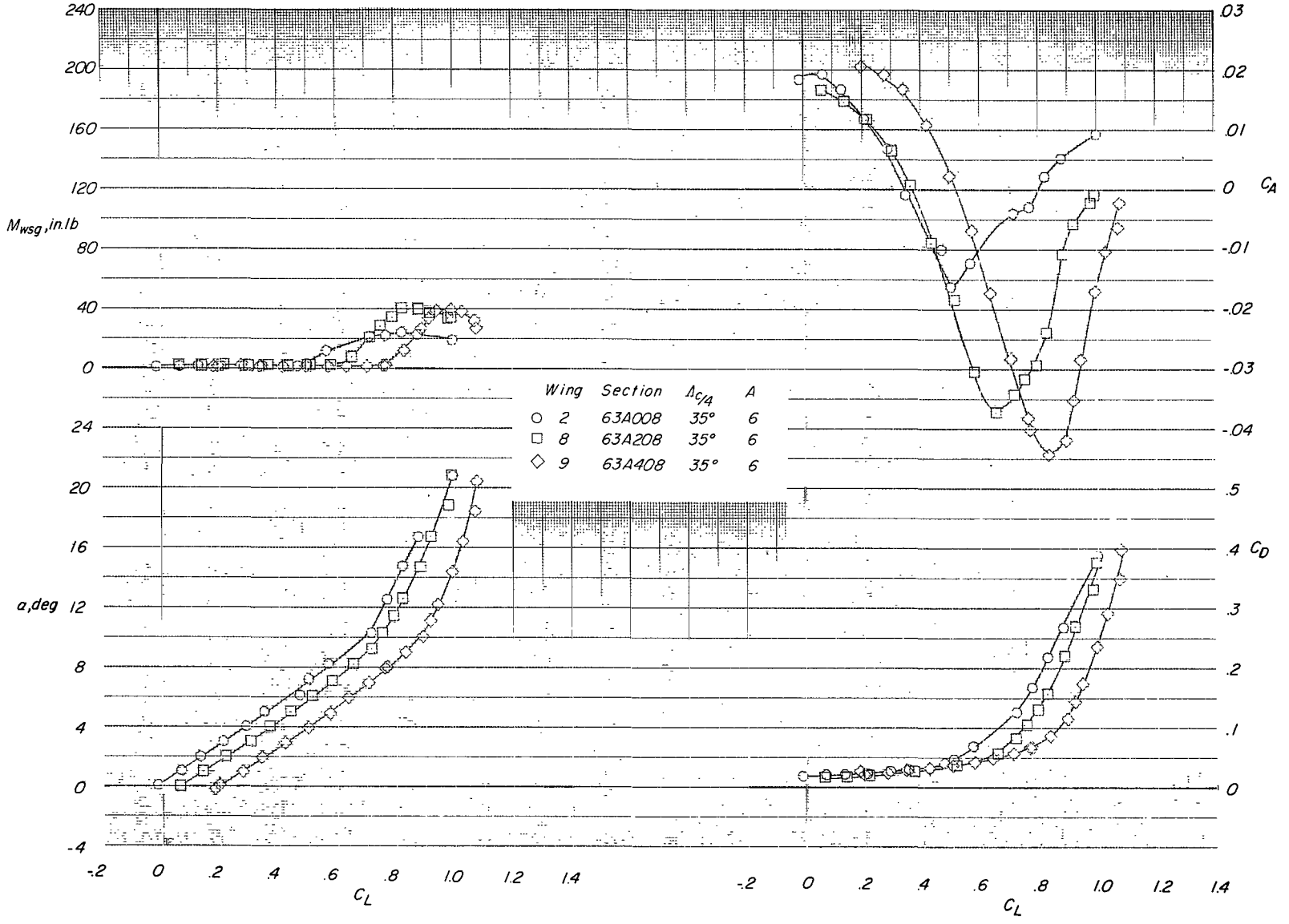
(e) Mach number, 0.82 for wing 2; 0.81 for wings 6 and 7.

Figure 20.- Continued.



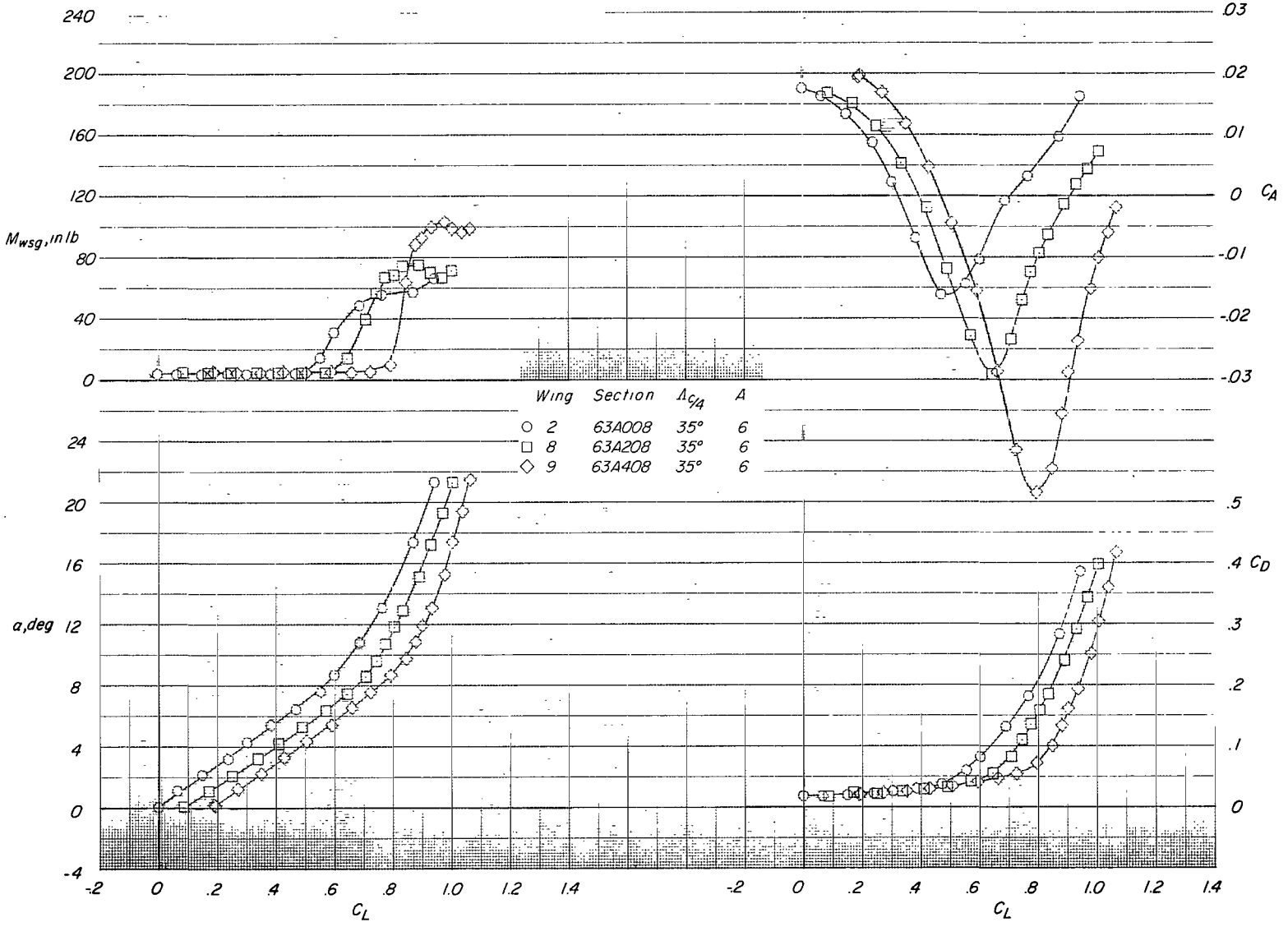
(f) Mach number, 0.88 for wing 2; 0.86 for wings 6 and 7.

Figure 20.- Concluded.



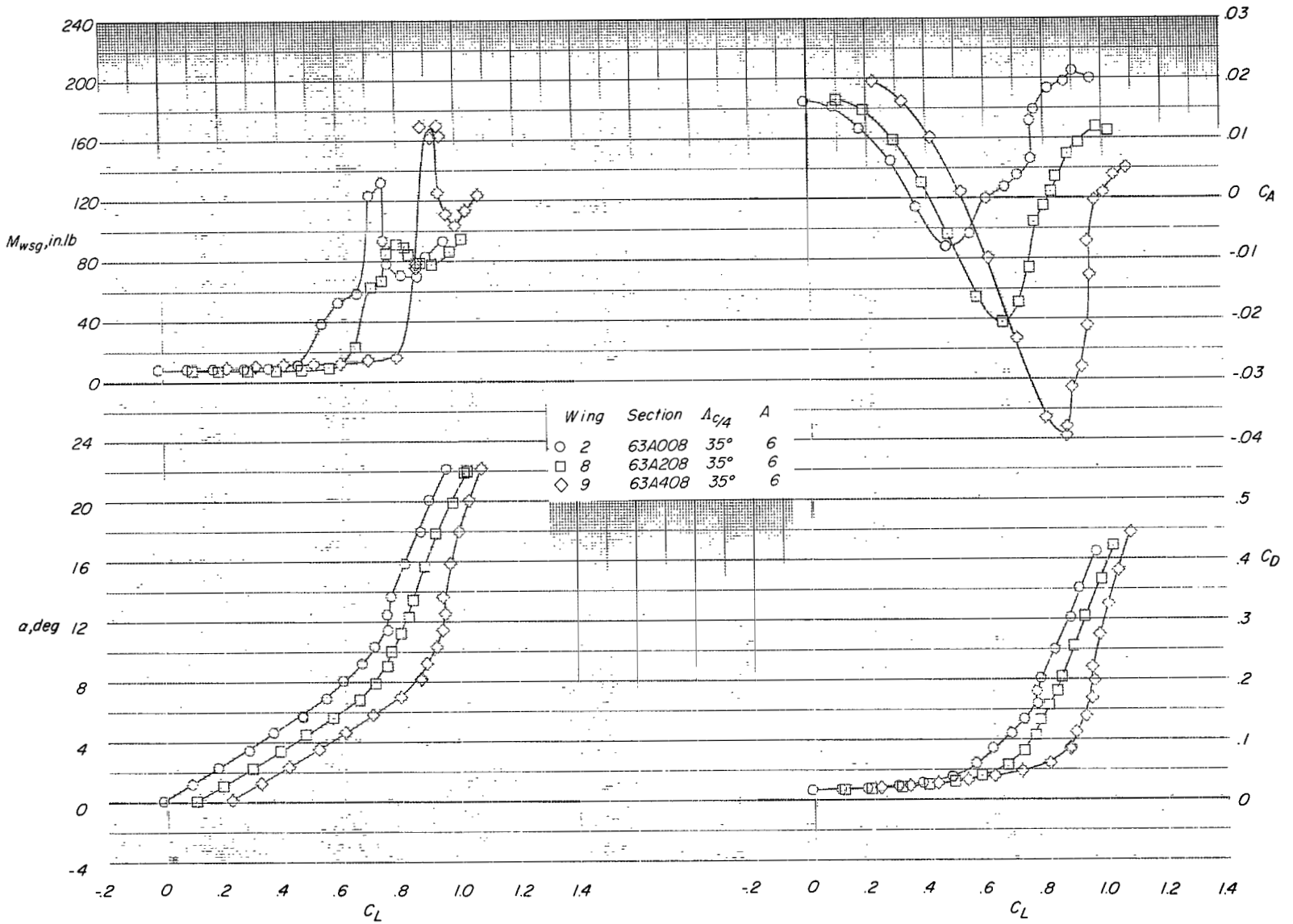
(a) Mach number, 0.23 for wings 2 and 9; 0.29 for wing 8.

Figure 21.- Effect of camber on the variations of M_{wsg}, α, C_A, and C_D with C_L. (1 in. lb = 0.113 m-N.)



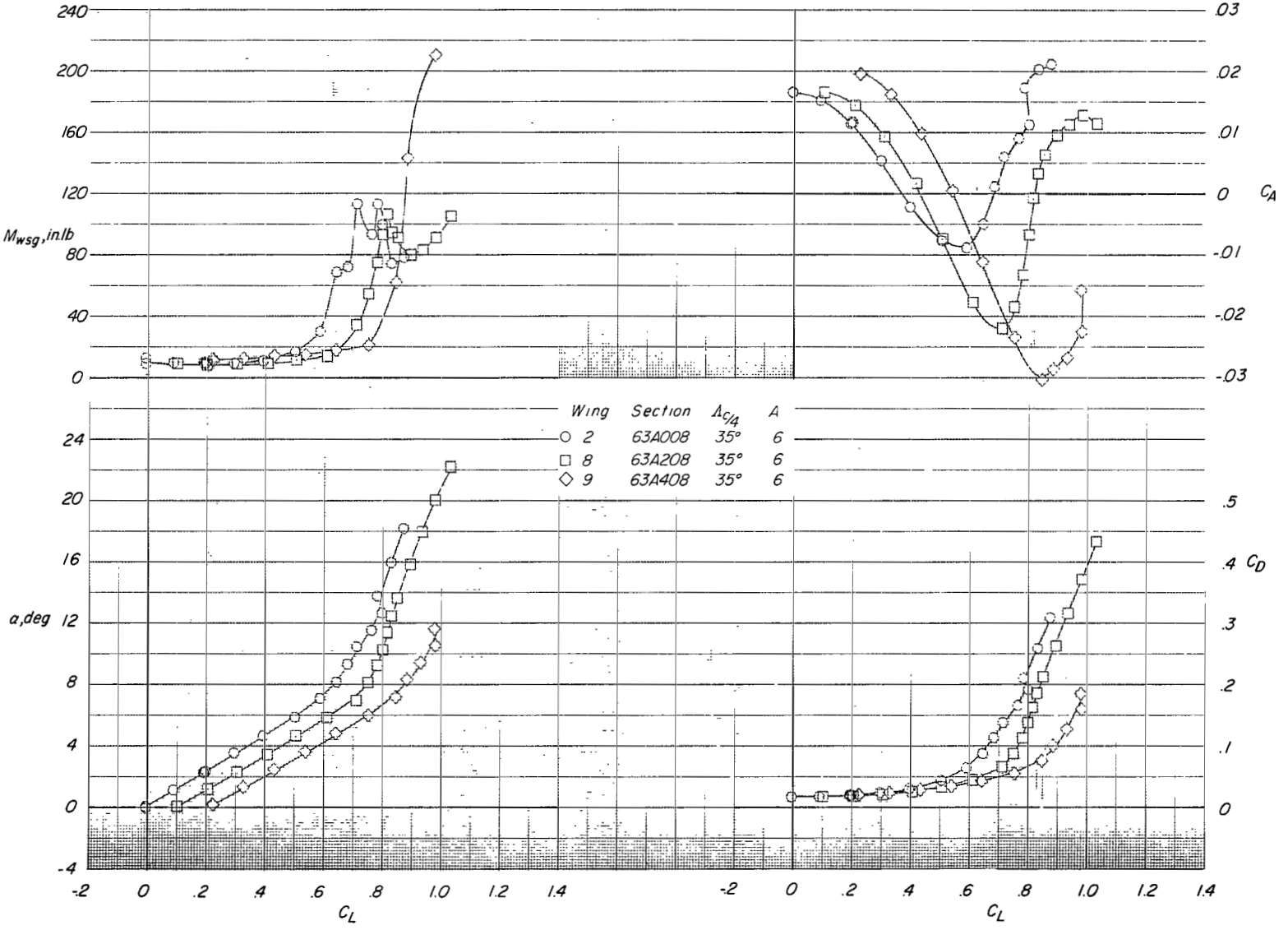
(b) Mach number, 0.46 for wings 2 and 9; 0.50 for wing 8.

Figure 21.- Continued.



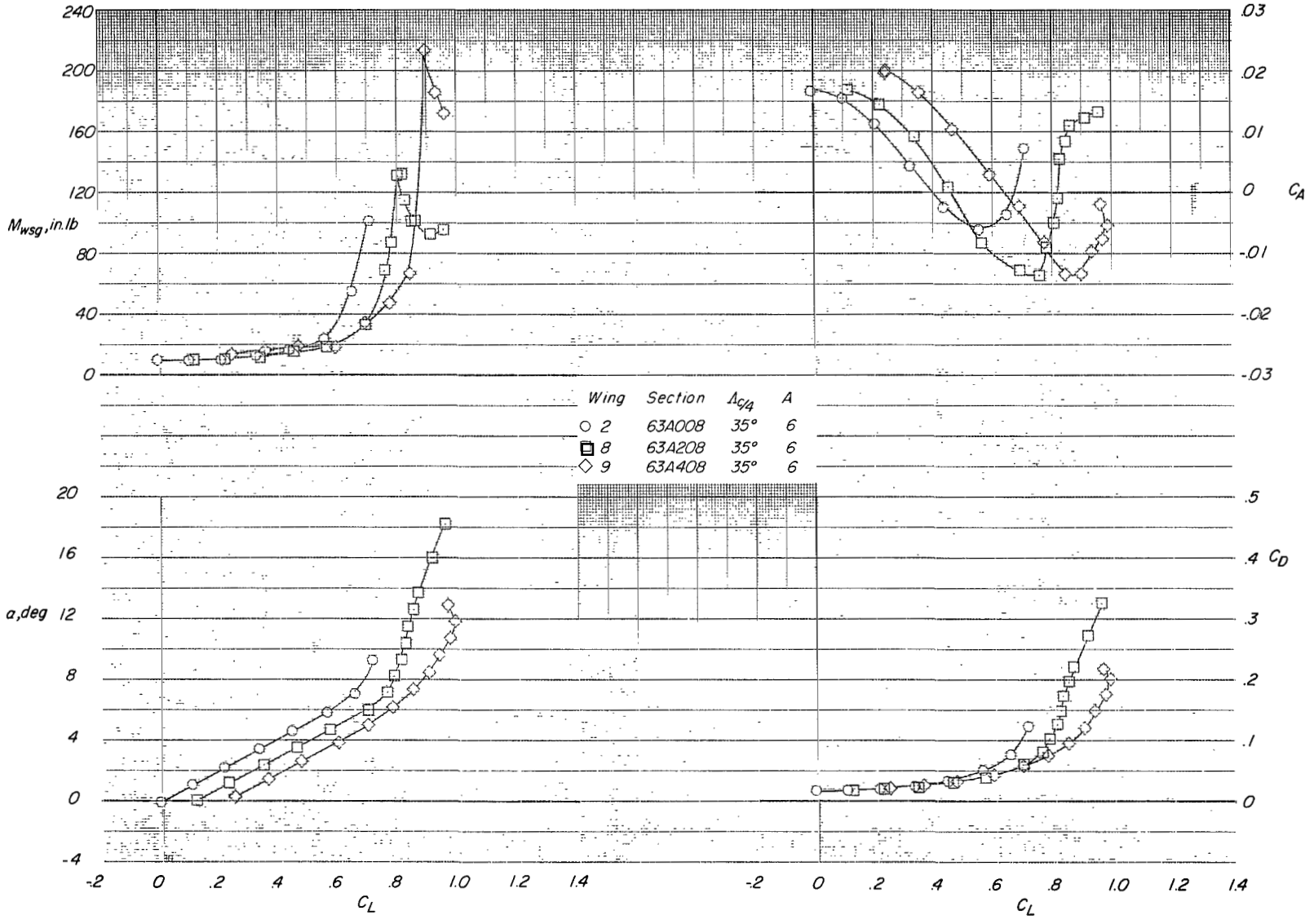
(c) Mach number, 0.72 for wings 2 and 9; 0.70 for wing 8.

Figure 21.- Continued.



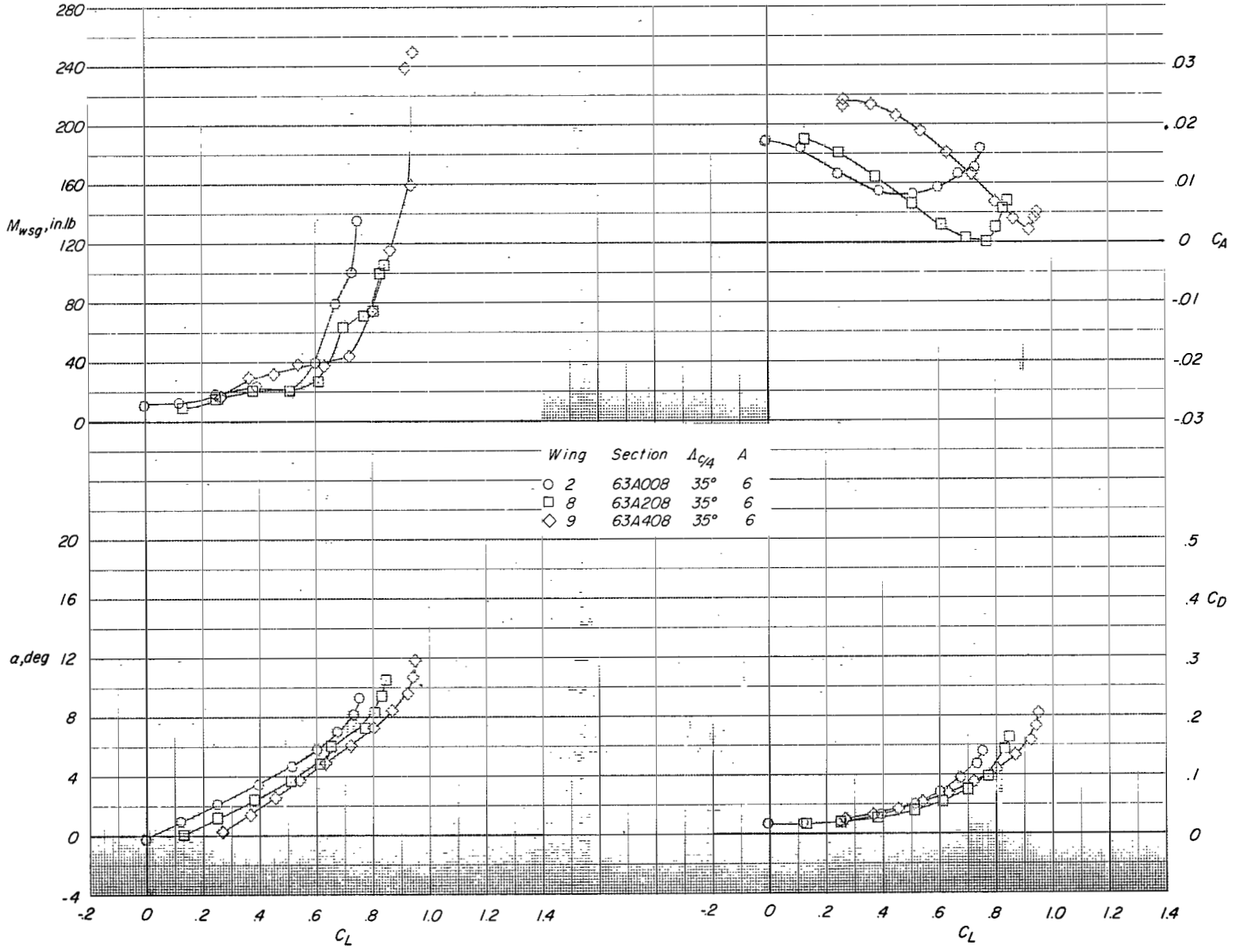
(d) Mach number, 0.77 for wings 2 and 9; 0.75 for wing 8.

Figure 21.- Continued.



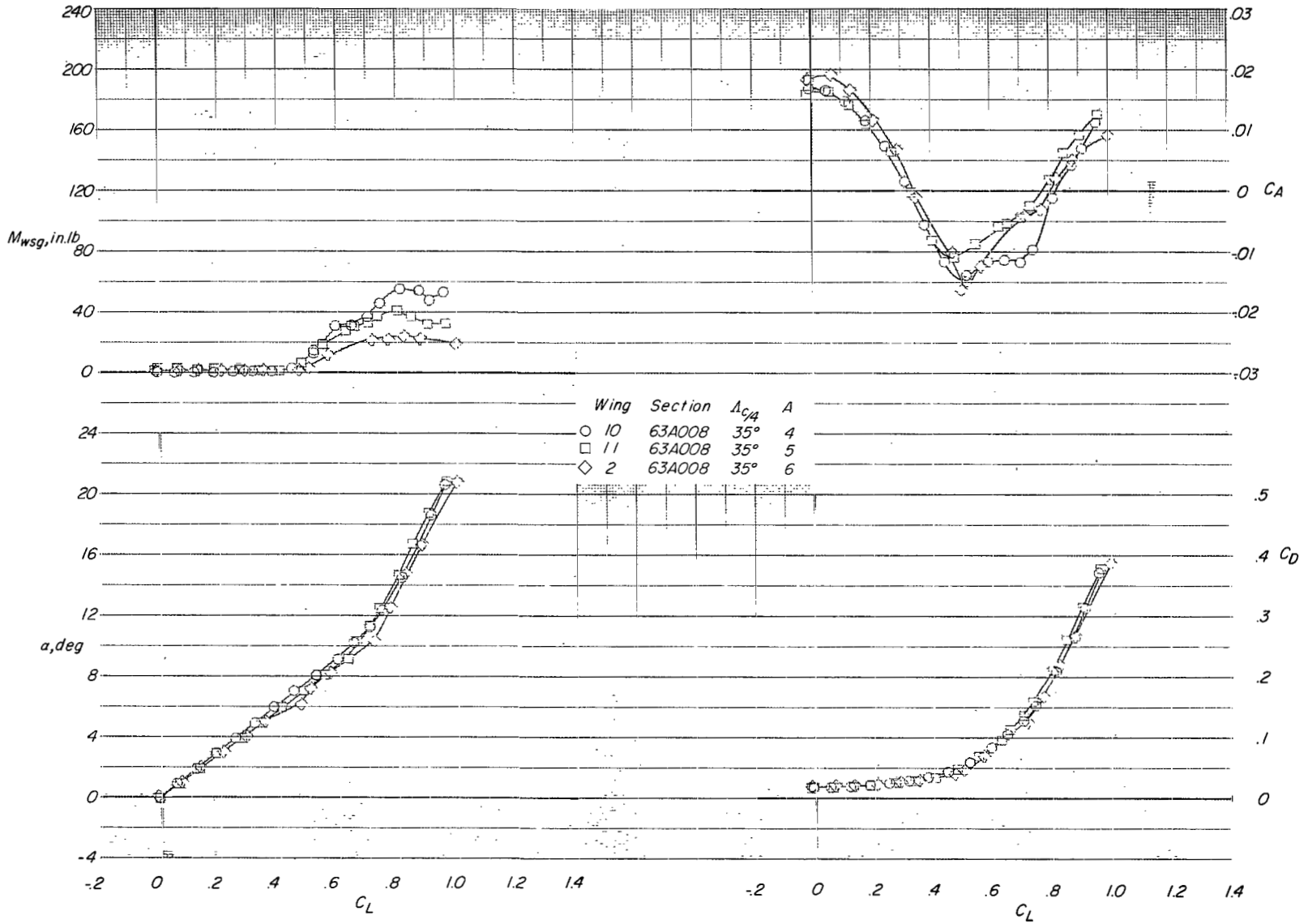
(e) Mach number, 0.82 for wings 2 and 9; 0.81 for wing 8.

Figure 21.- Continued.



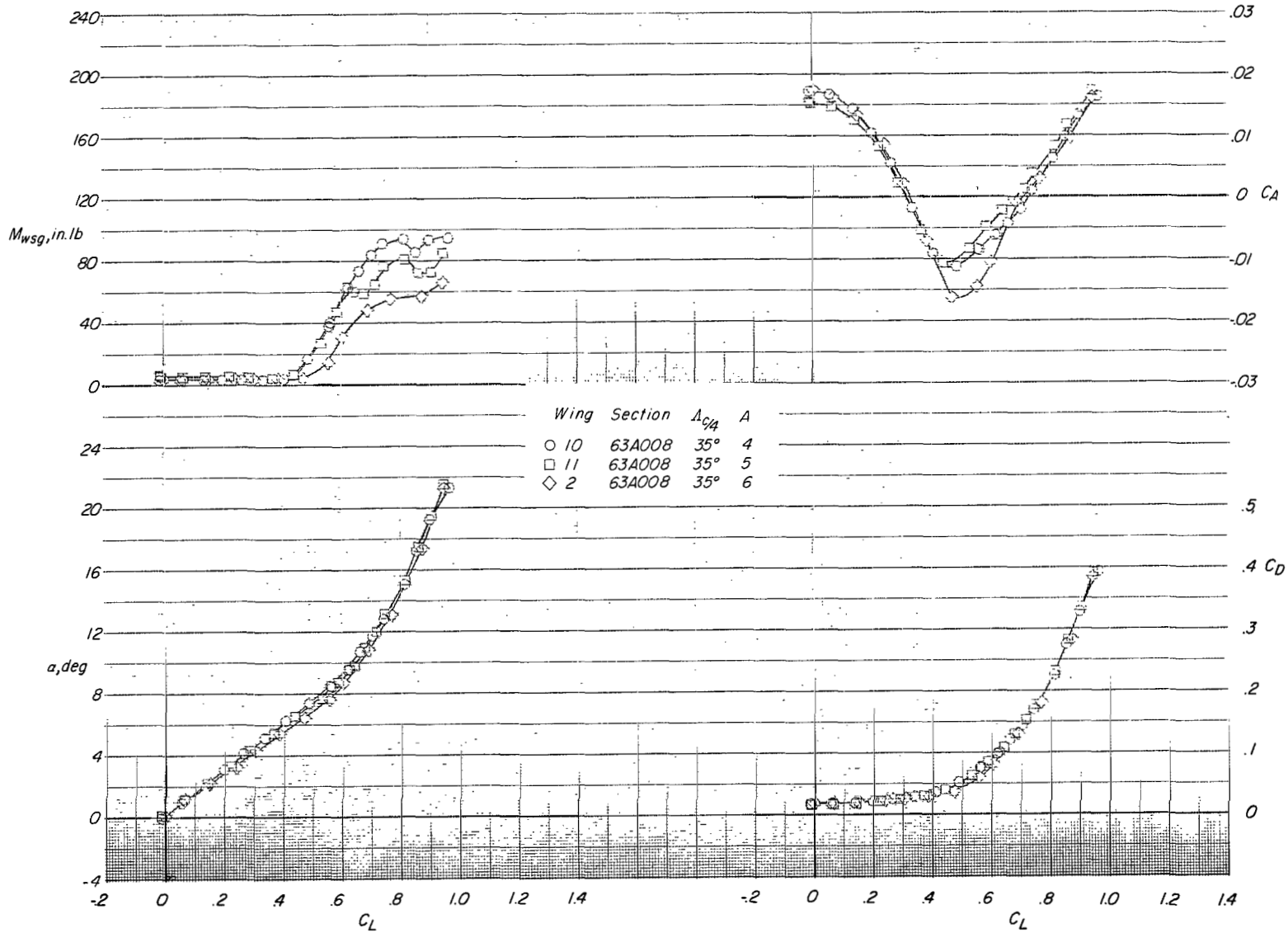
(f) Mach number, 0.88 for wings 2 and 9; 0.86 for wing 8.

Figure 21.- Concluded.



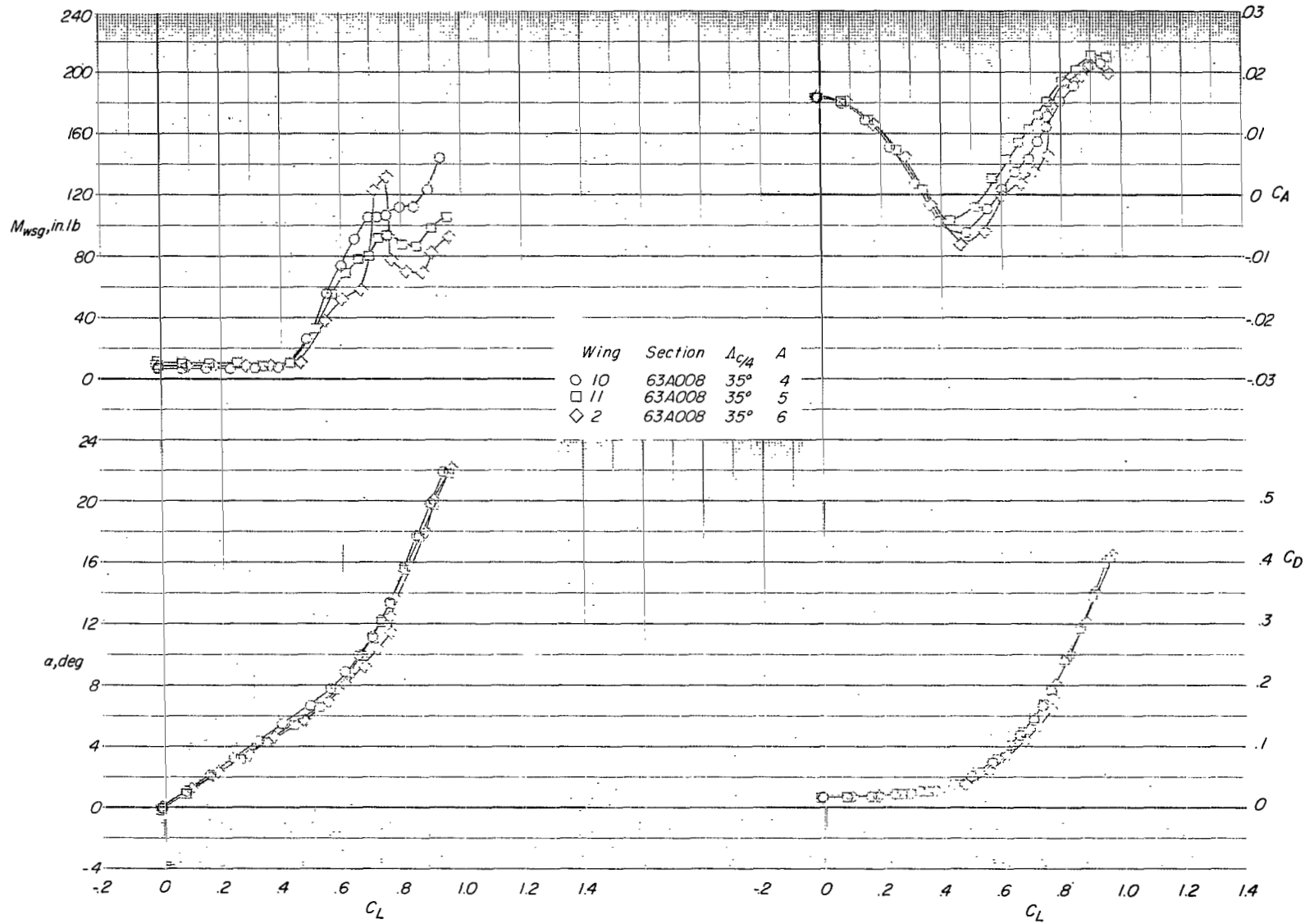
(a) Mach number, 0.23 for wing 2; 0.29 for wings 10 and 11.

Figure 22.- Effect of aspect ratio on the variations of M_{wsg}, α, C_A, and C_D with C_L. (1 in. lb = 0.113 m-N.)



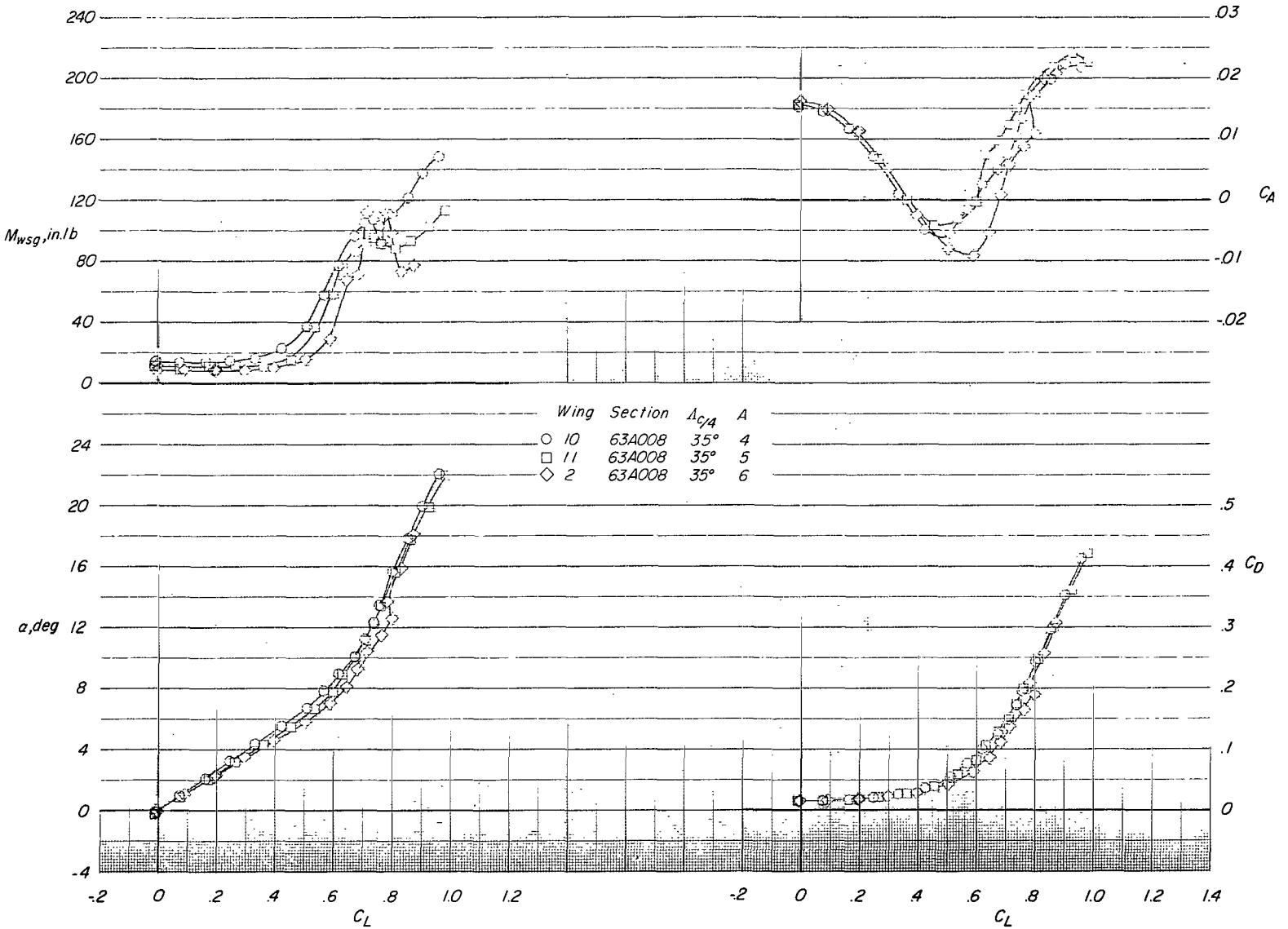
(b) Mach number, 0.46 for wing 2; 0.50 for wings 10 and 11.

Figure 22.- Continued.



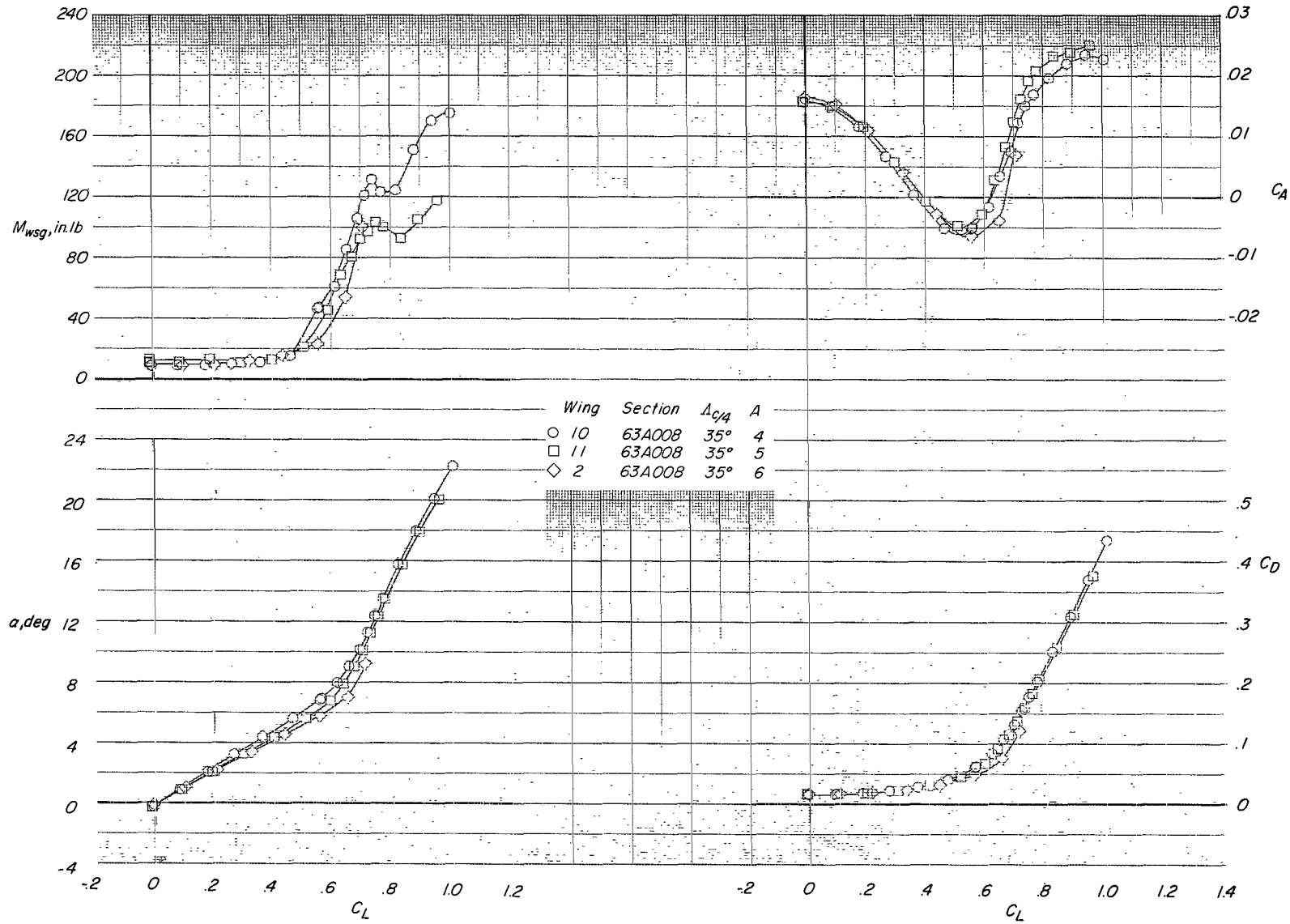
(c) Mach number, 0.72 for wing 2; 0.70 for wings 10 and 11.

Figure 22.- Continued.



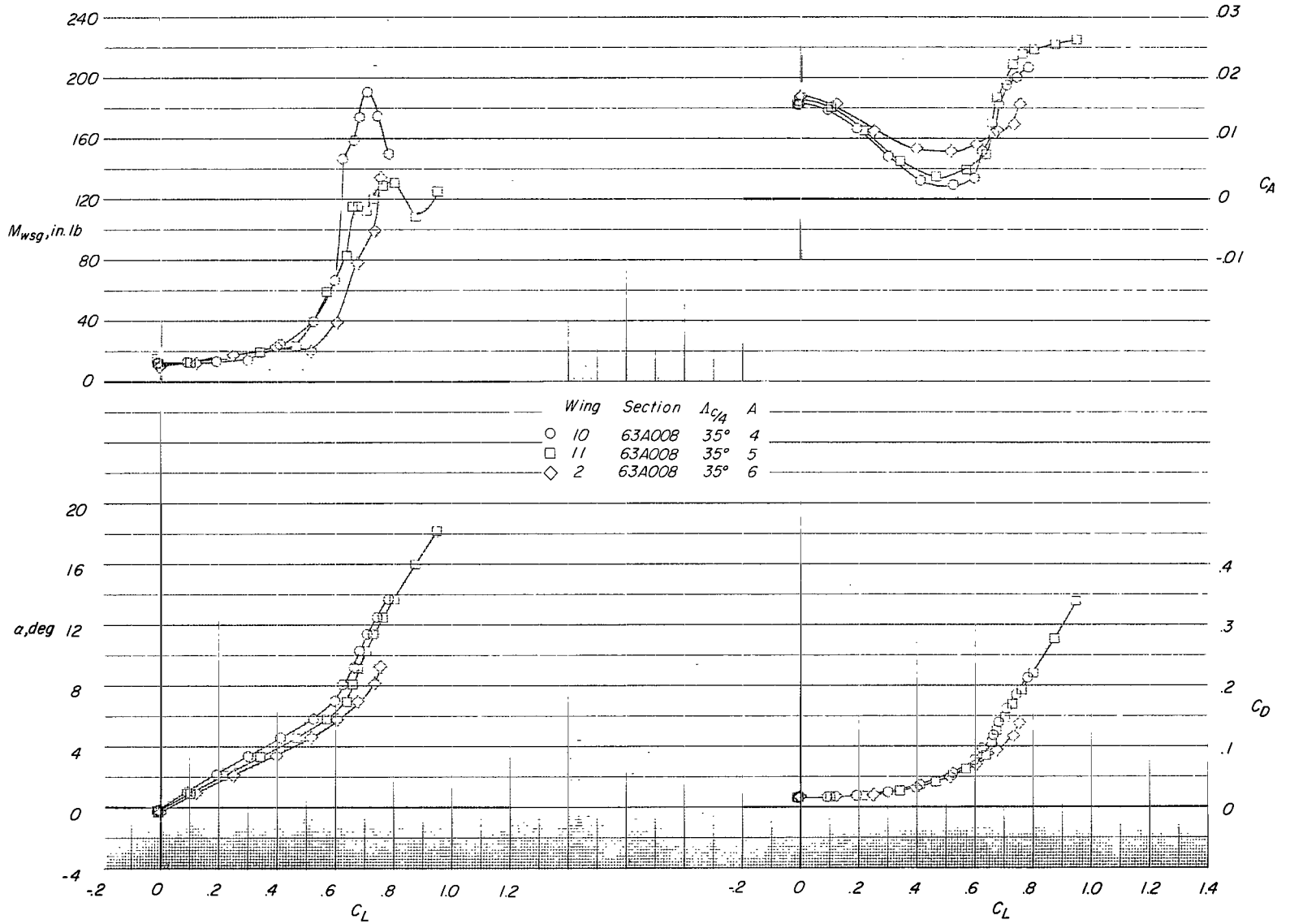
(d) Mach number, 0.77 for wing 2; 0.75 for wings 10 and 11.

Figure 22.- Continued.



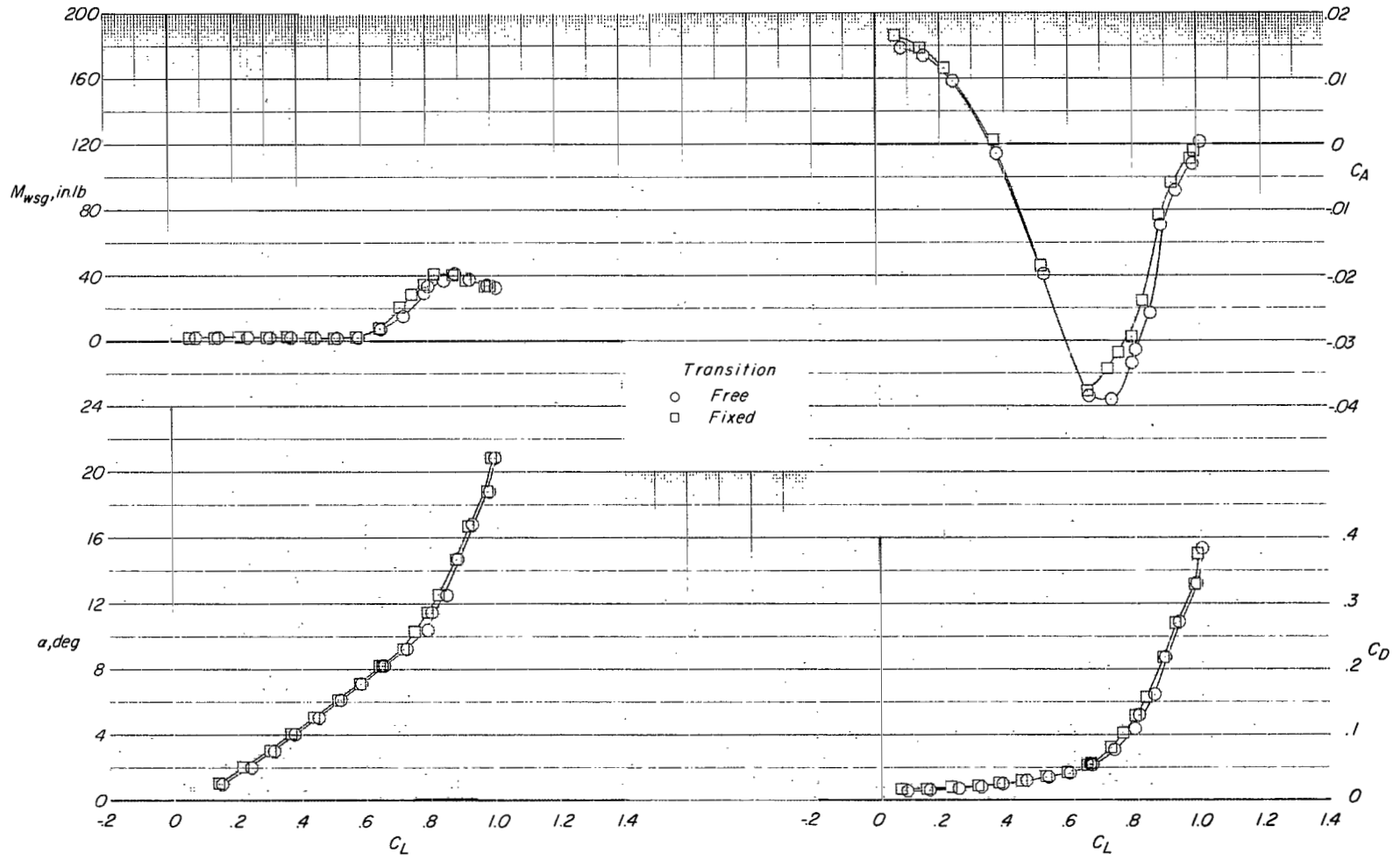
(e) Mach number, 0.82 for wing 2; 0.81 for wings 10 and 11.

Figure 22.- Continued.



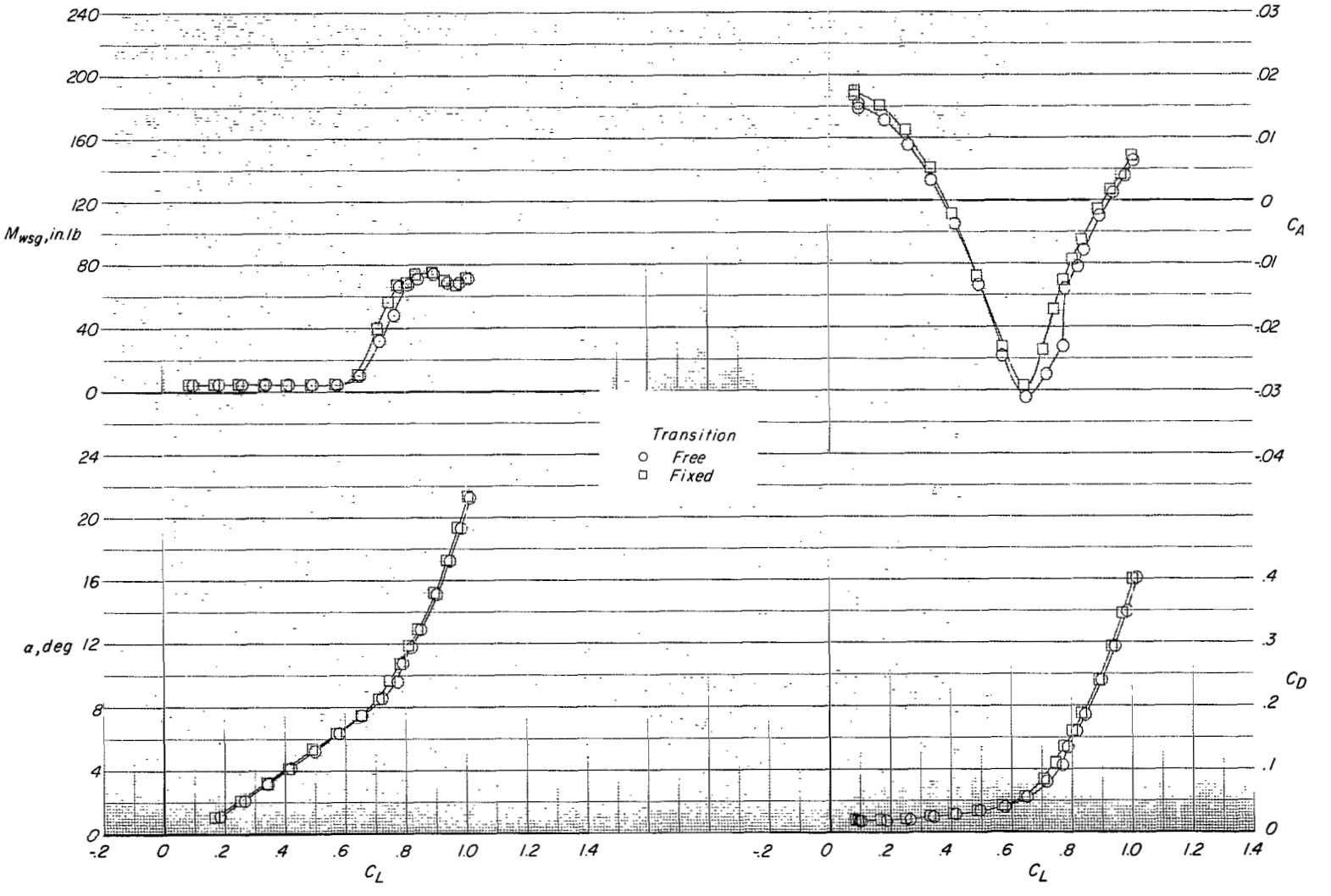
(f) Mach number, 0.88 for wing 2; 0.86 for wings 10 and 11.

Figure 22.- Concluded.



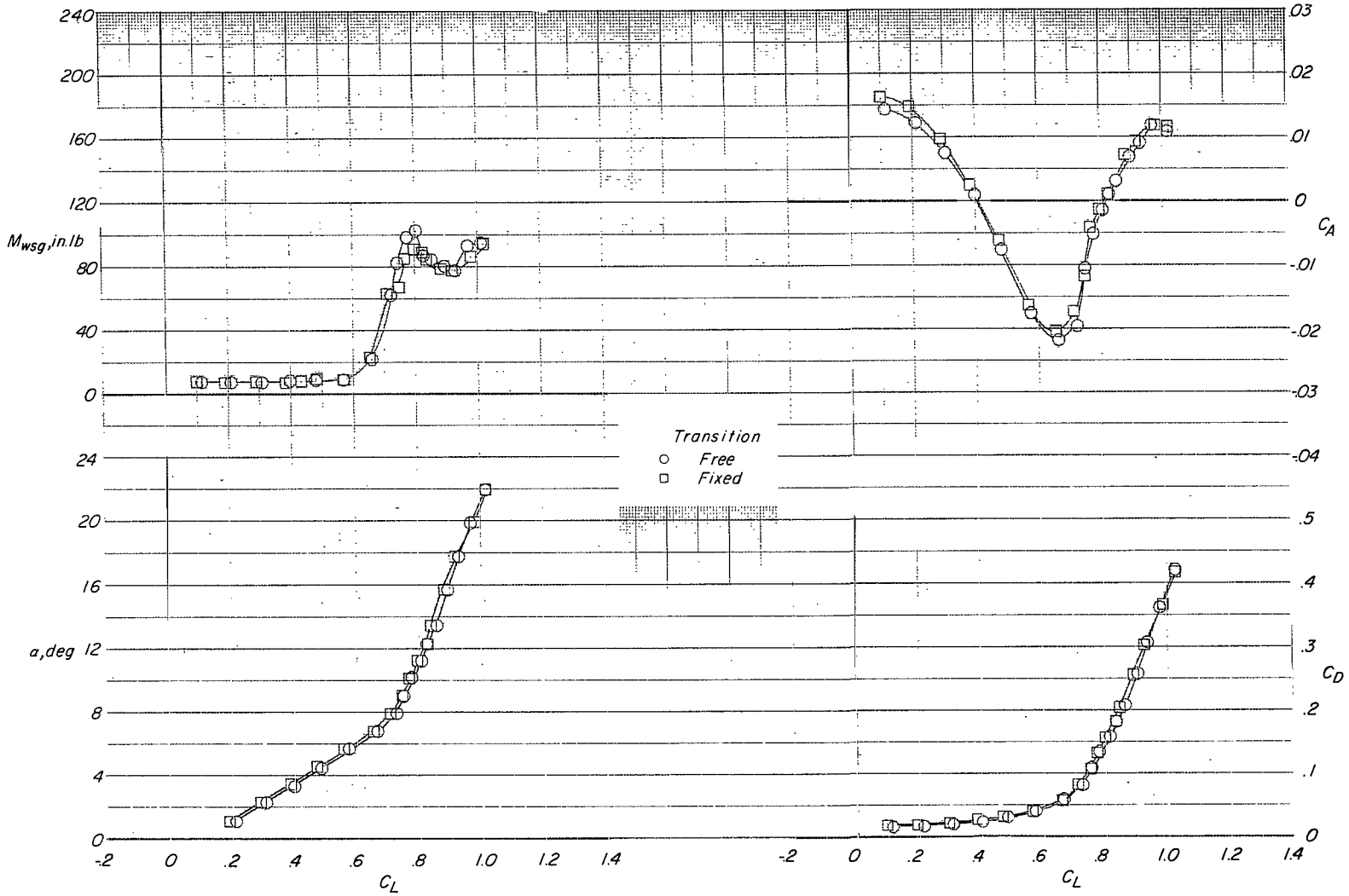
(a) Mach number, 0.29.

Figure 23.- Effect of transition grit on the variations of M_{wsg}, α, C_A, and C_D with C_L for wing 8. (1 in. lb = 0.113 m-N.)



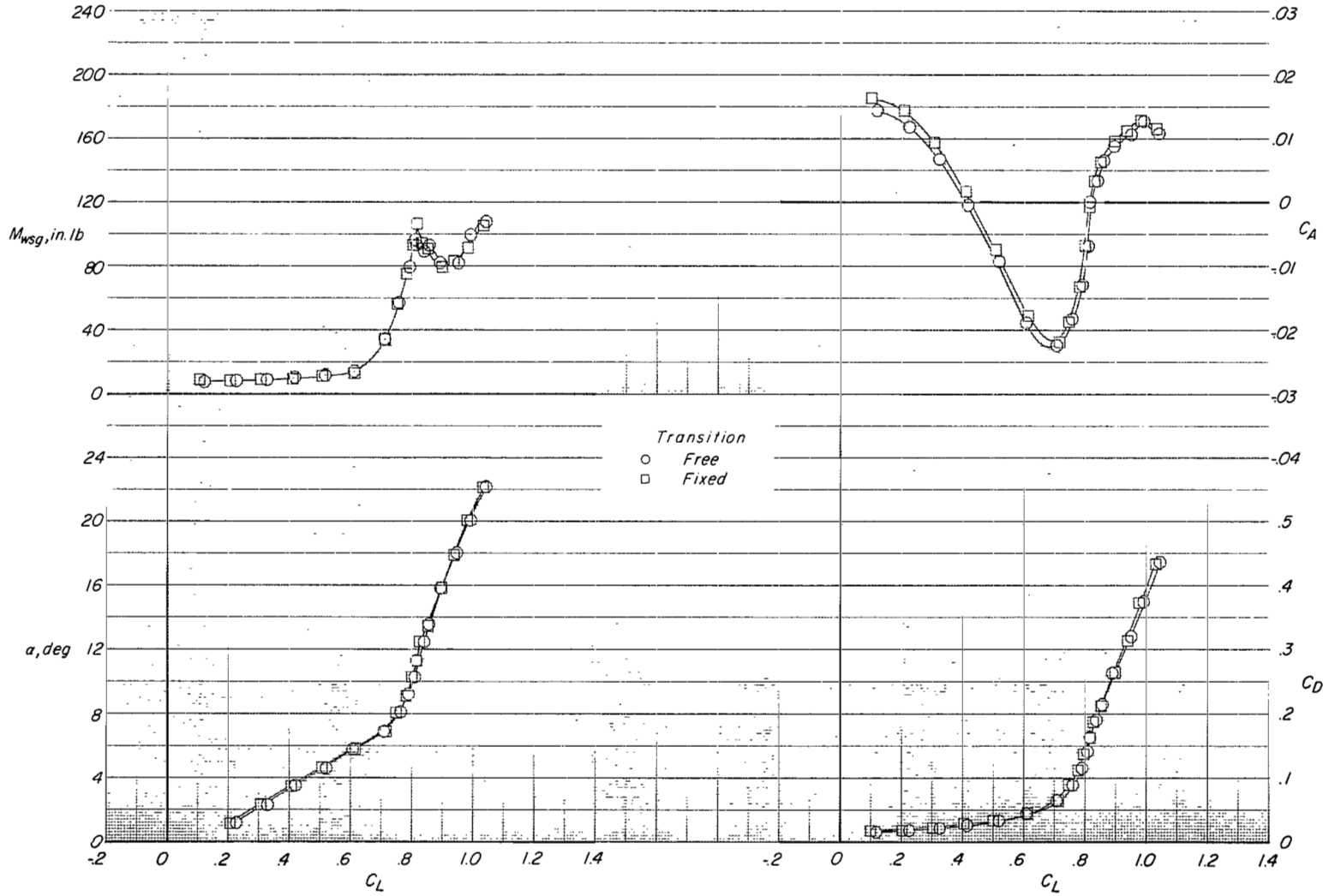
(b) Mach number, 0.50.

Figure 23.- Continued.



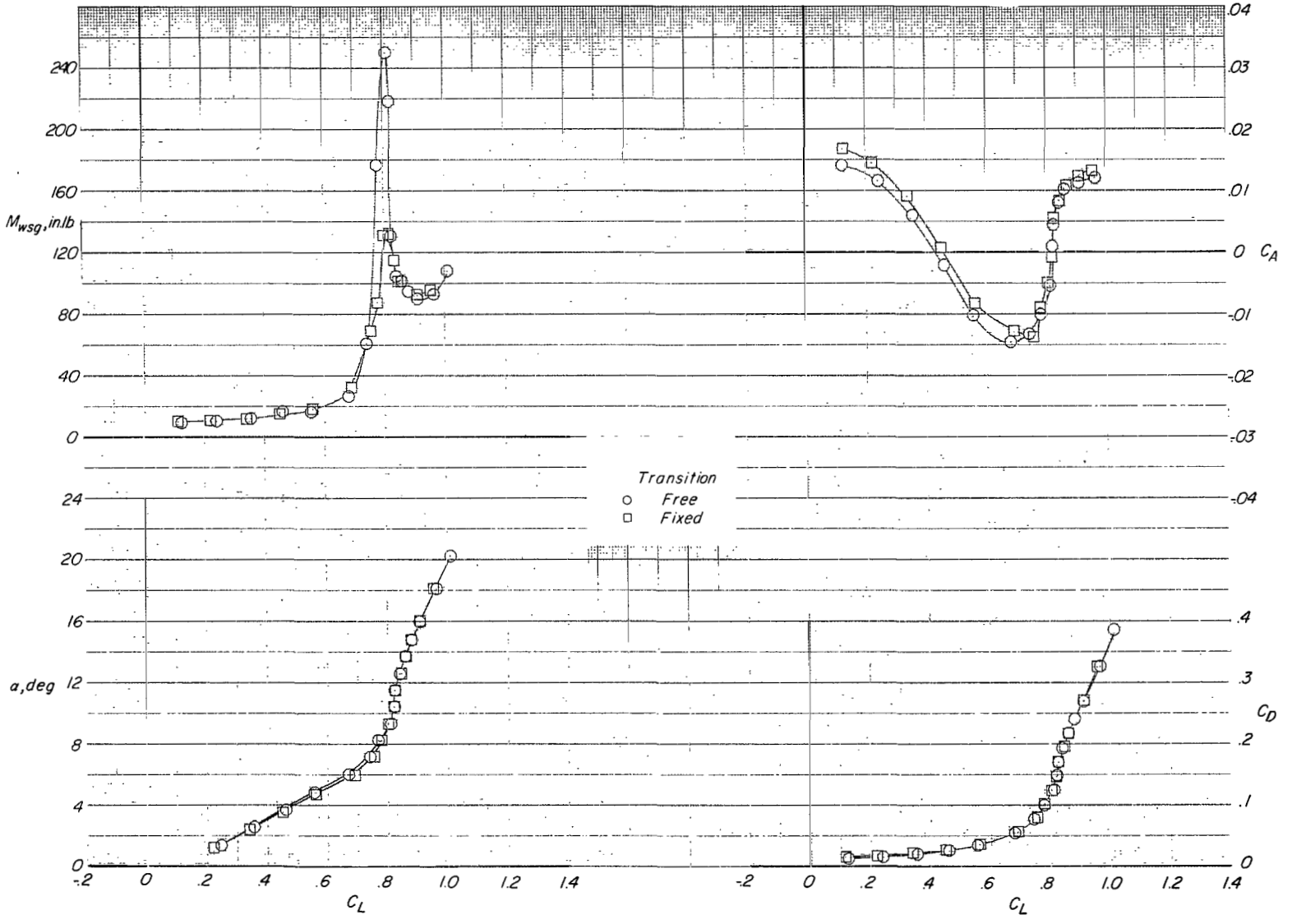
(c) Mach number, 0.70.

Figure 23.- Continued.



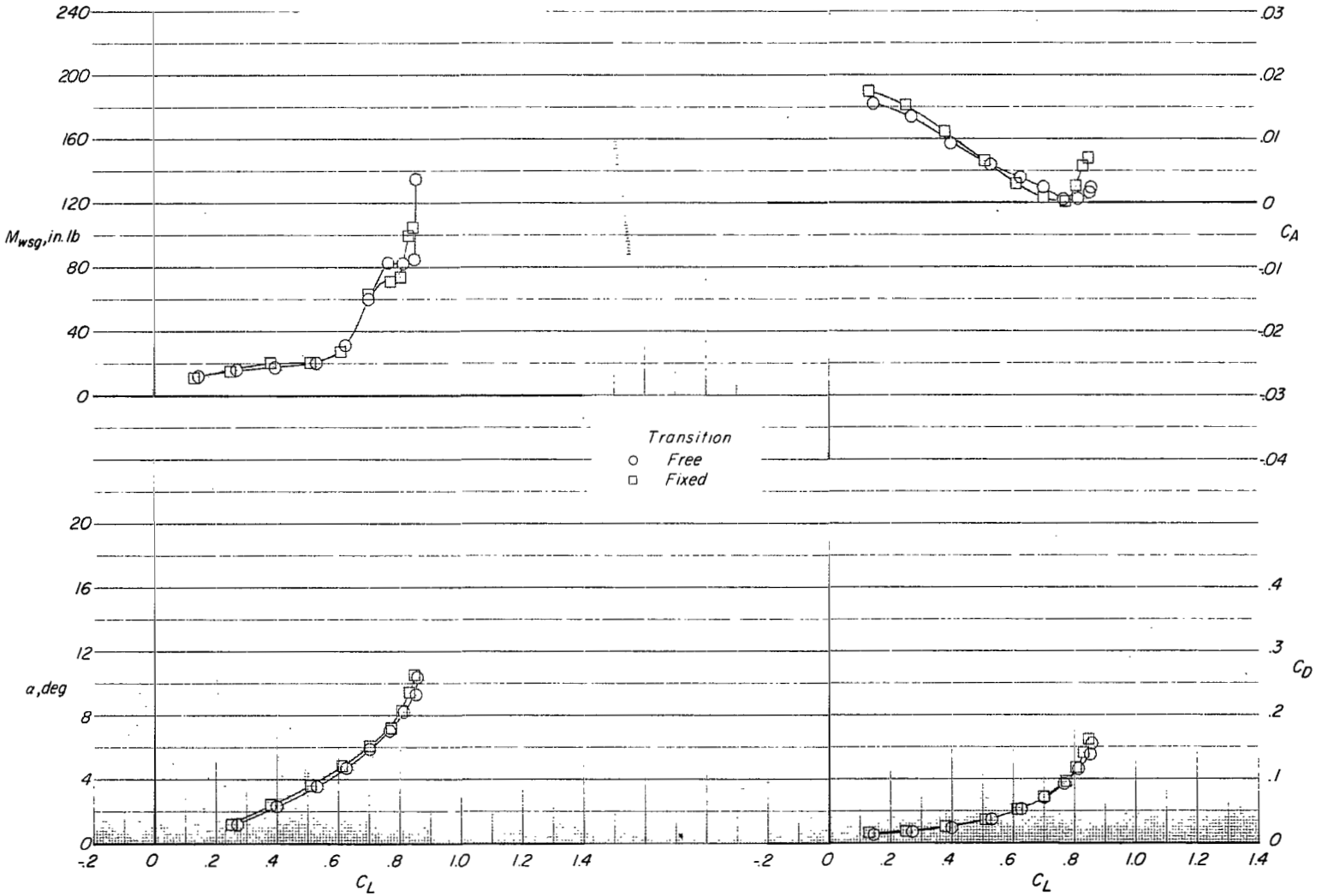
(d) Mach number, 0.75.

Figure 23.- Continued.



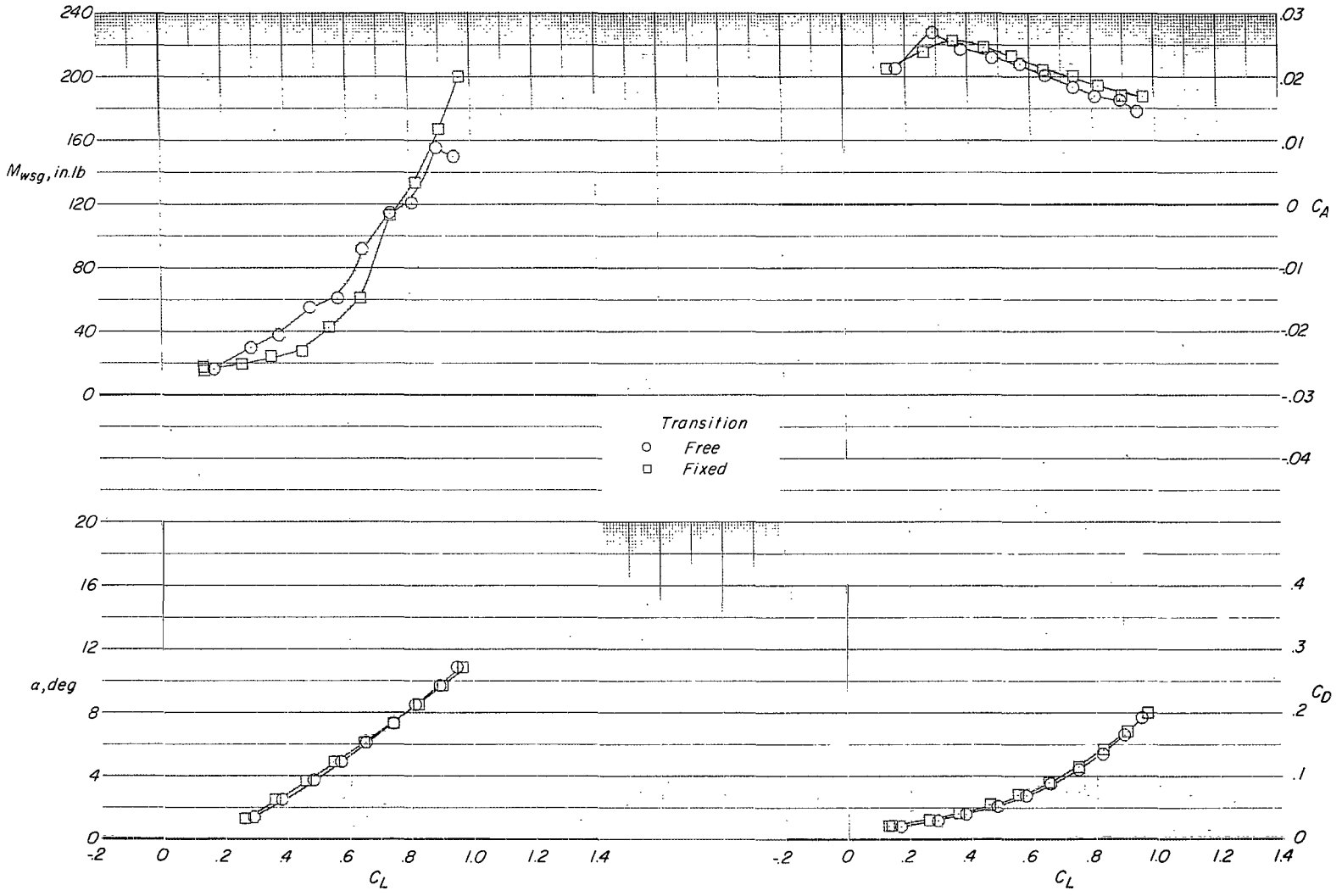
(e) Mach number, 0.81.

Figure 23.- Continued.



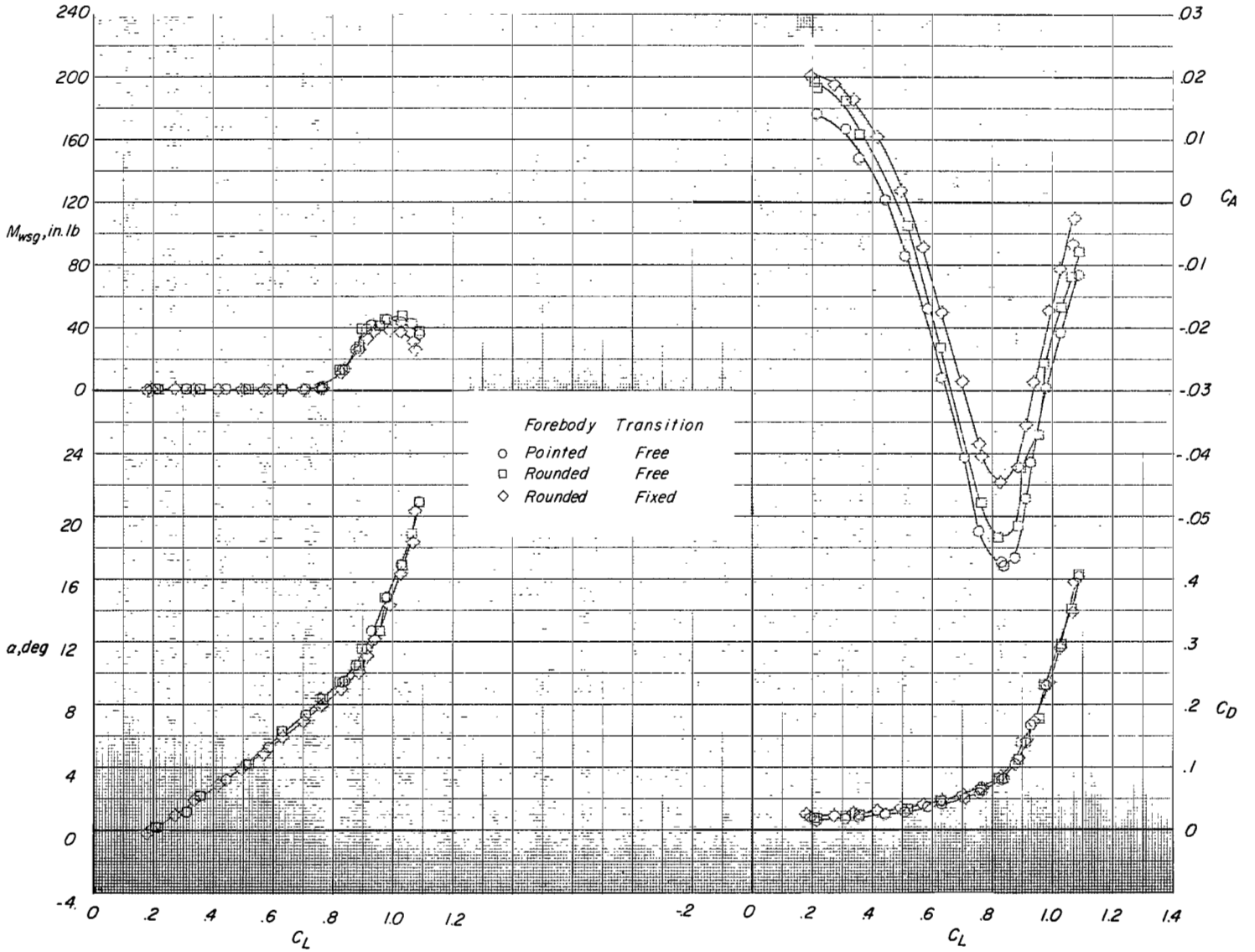
(f) Mach number, 0.86.

Figure 23.- Continued.



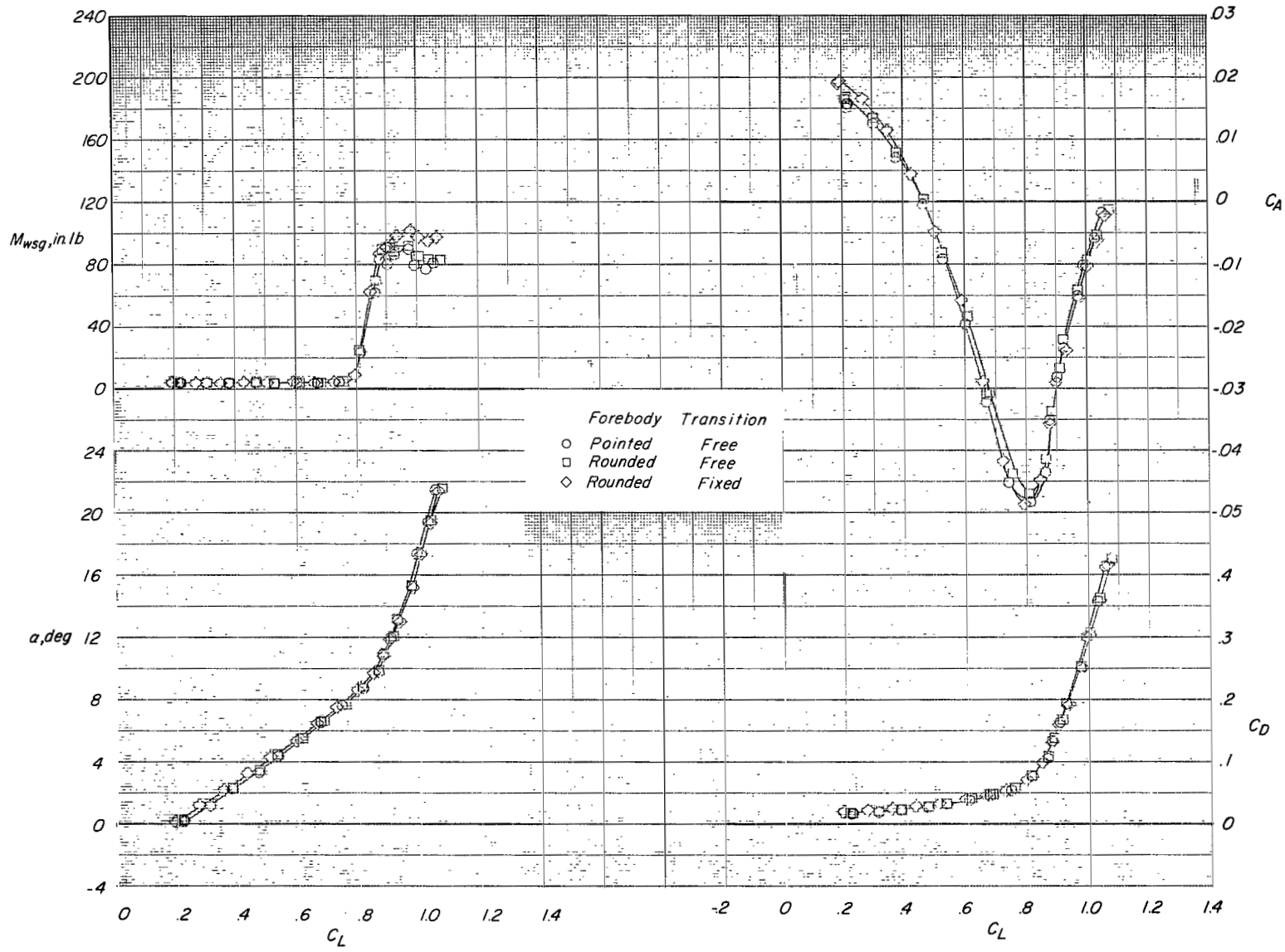
(g) Mach number, 0.93.

Figure 23.- Concluded.



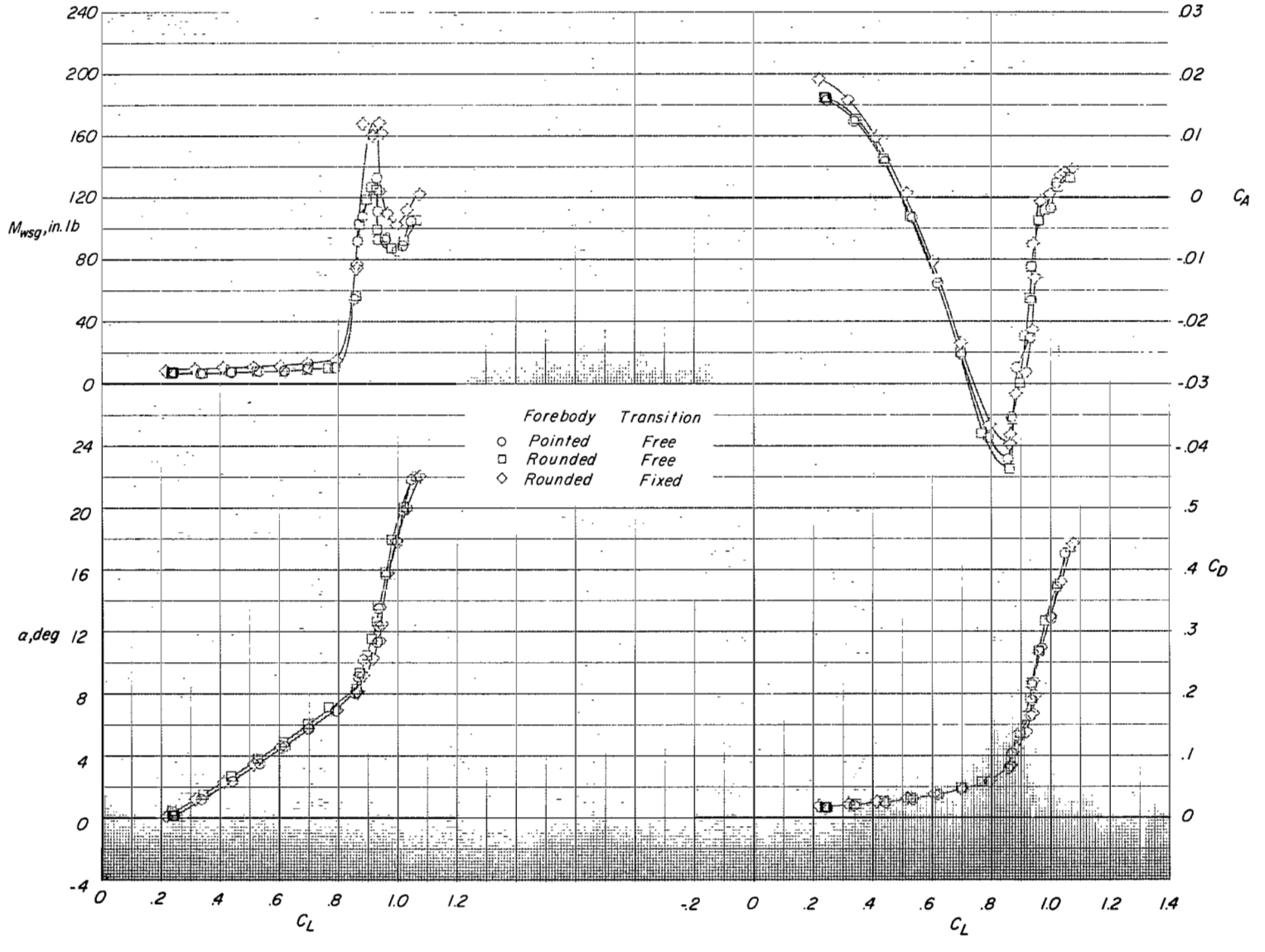
(a) Mach number, 0.29 with grit off; 0.23 with grit on.

Figure 24.- Effect of forebody shape and transition grit on the variations of M_{wsg}, α, C_A, and C_D with C_L for wing 9. (1 in. lb = 0.113 m-N.)



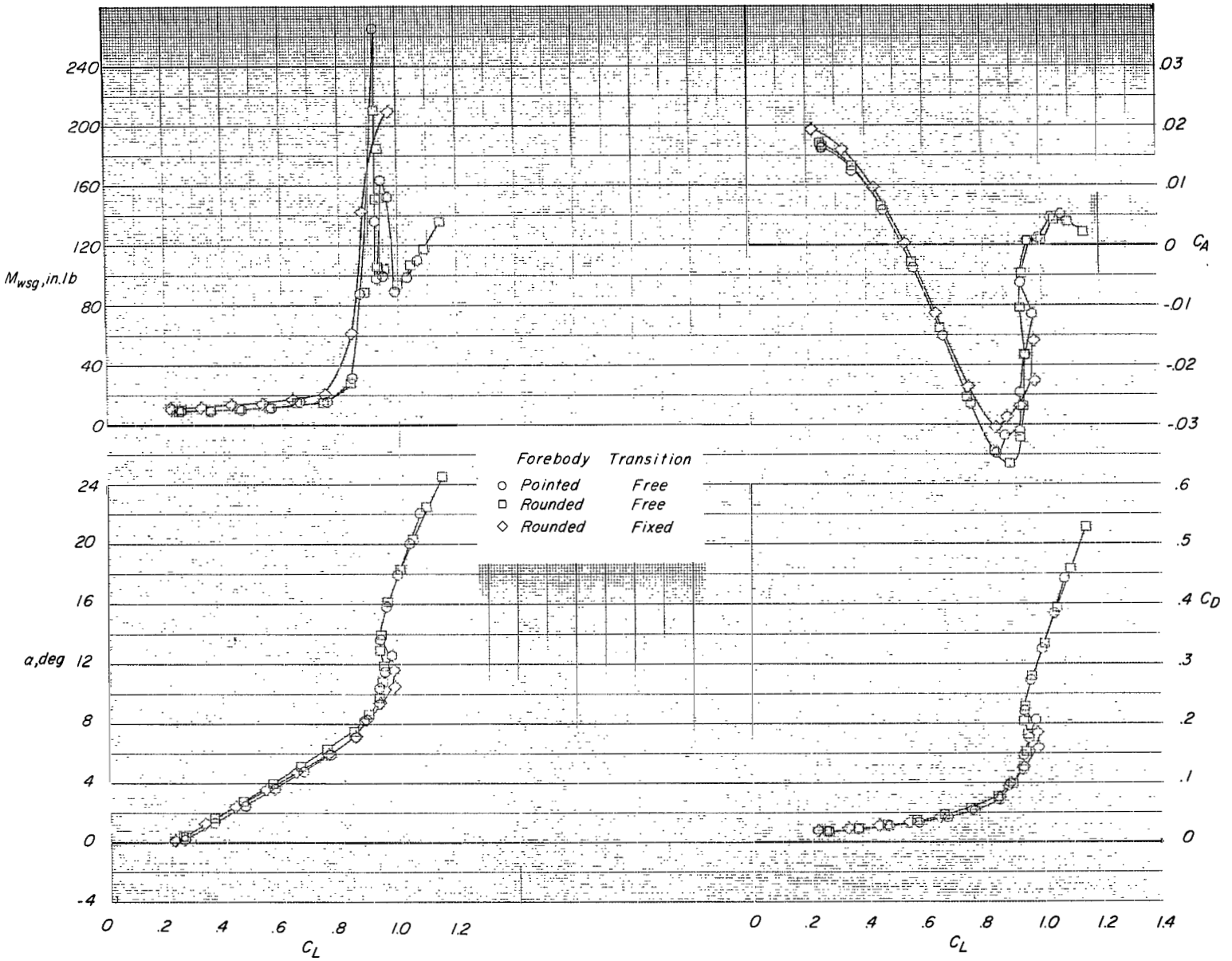
(b) Mach number, 0.50 with grit off; 0.46 with grit on.

Figure 24.- Continued.



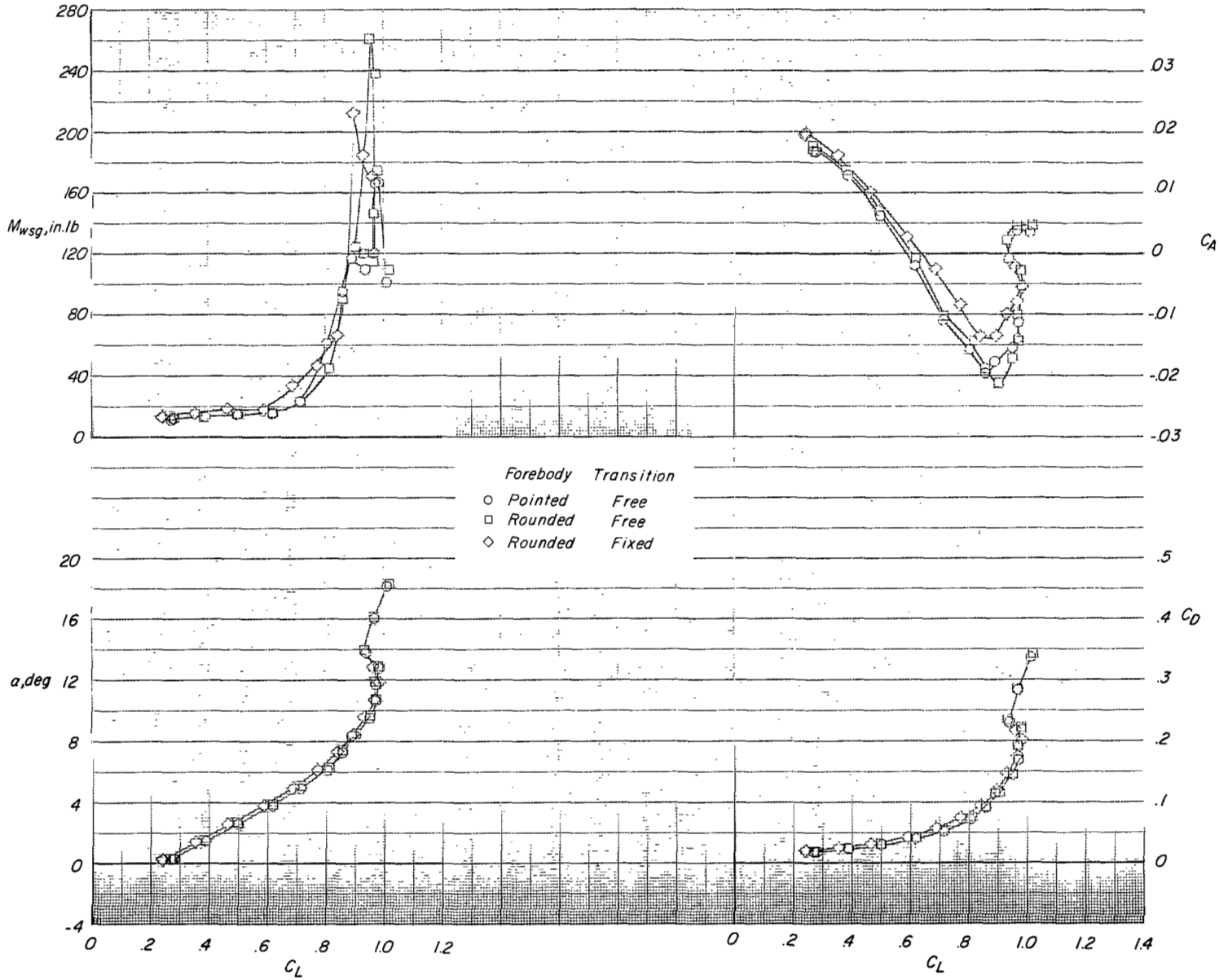
(c) Mach number, 0.70 with grit off; 0.72 with grit on.

Figure 24.- Continued.



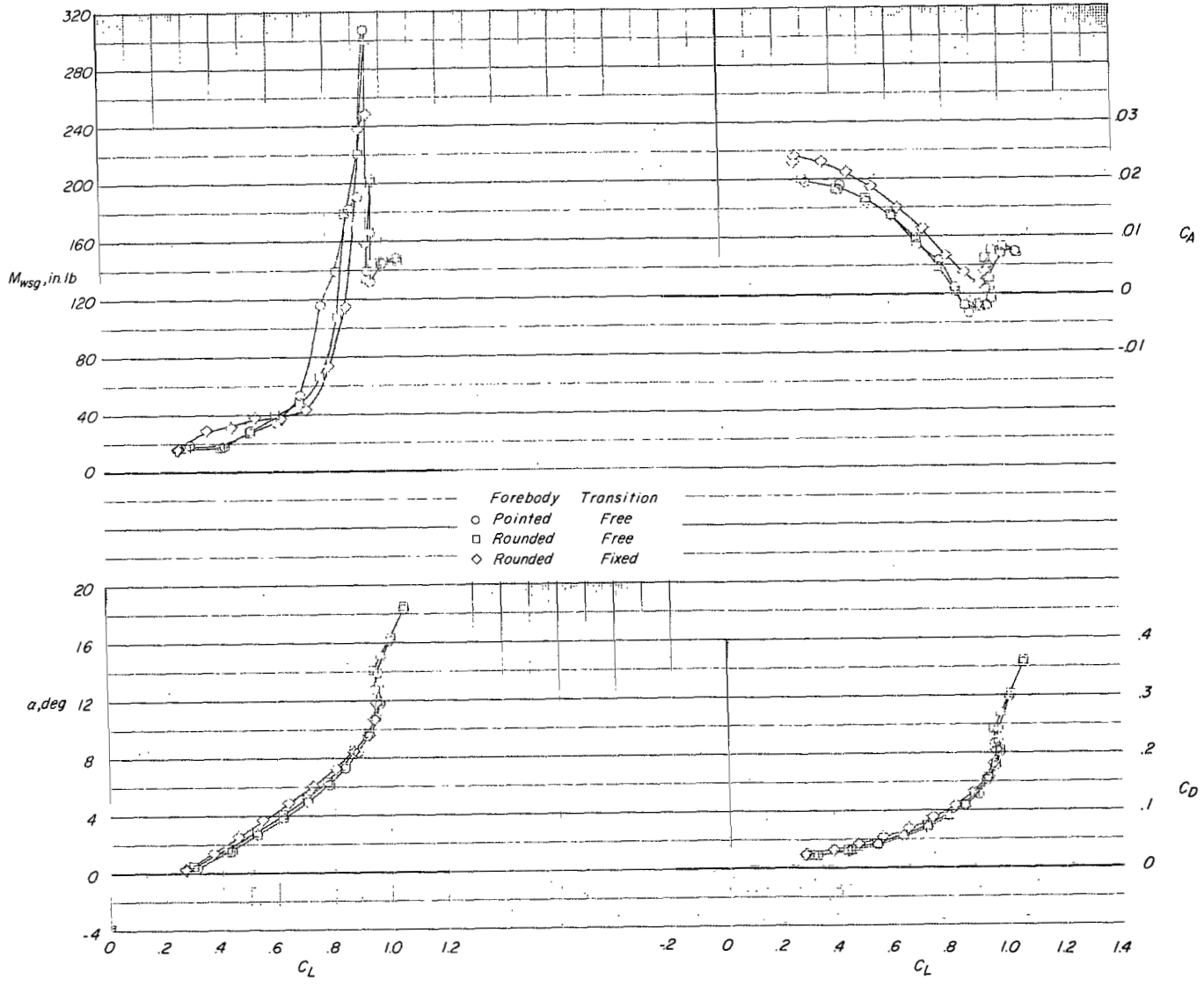
(d) Mach number, 0.75 with grit off; 0.77 with grit on.

Figure 24.- Continued.



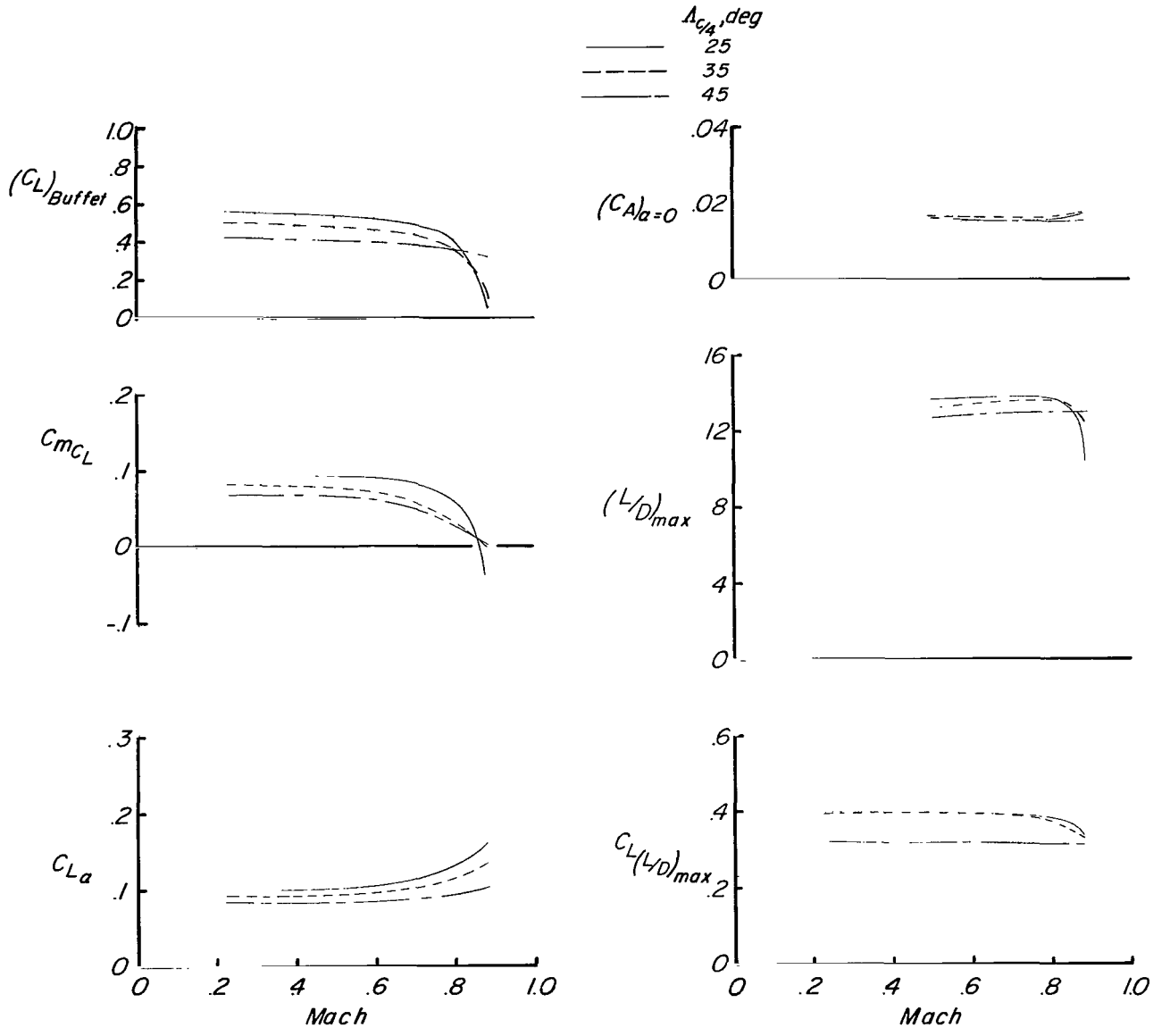
(e) Mach number, 0.81 with grit off; 0.82 with grit on.

Figure 24.- Continued.



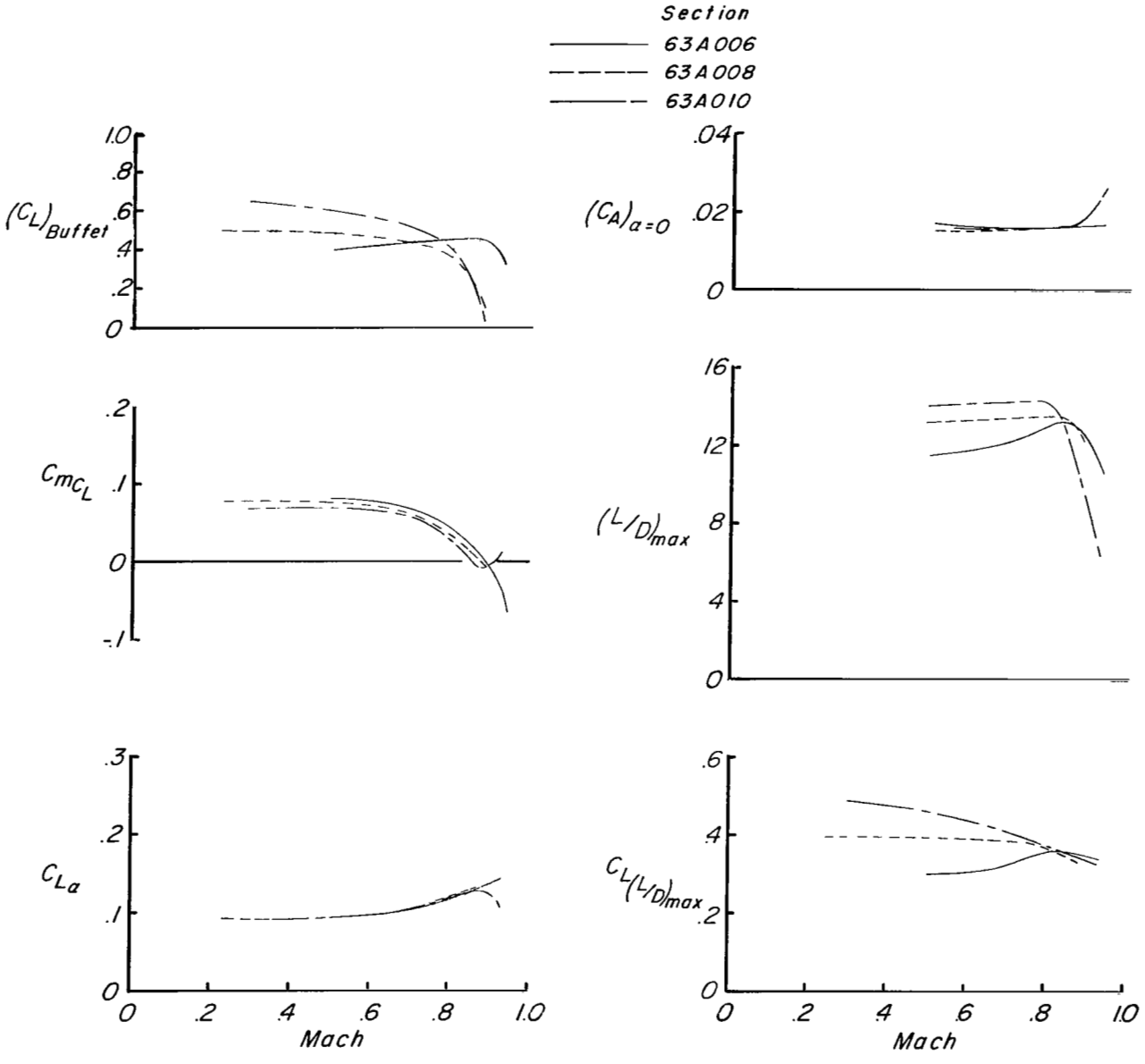
(f) Mach number, 0.86 with grit off; 0.88 with grit on.

Figure 24.- Concluded.



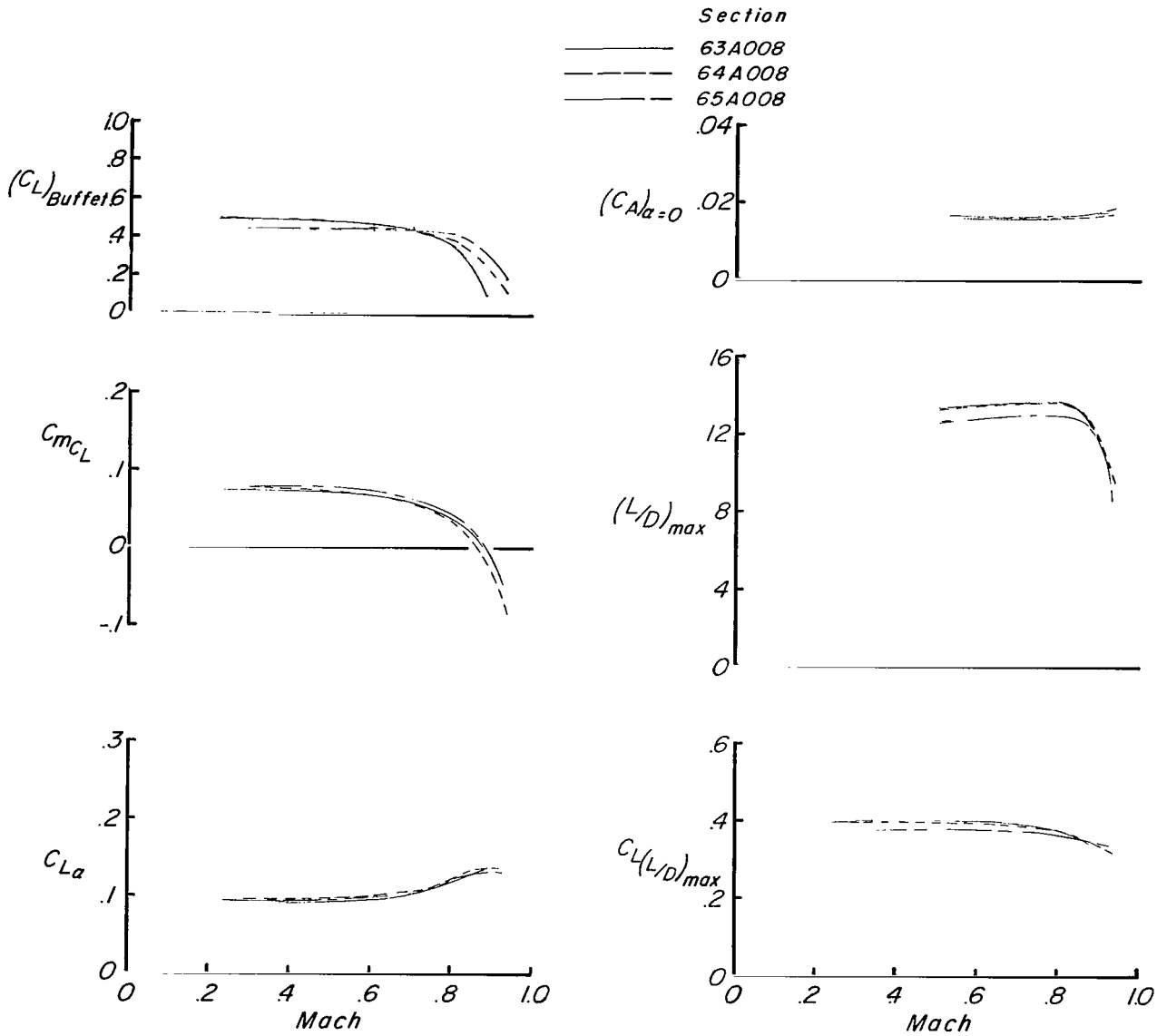
(a) Effects of wing sweep. Airfoil, NACA 63A008; A = 6.

Figure 25.- Summaries of the longitudinal aerodynamic and buffet characteristics of the test configurations.



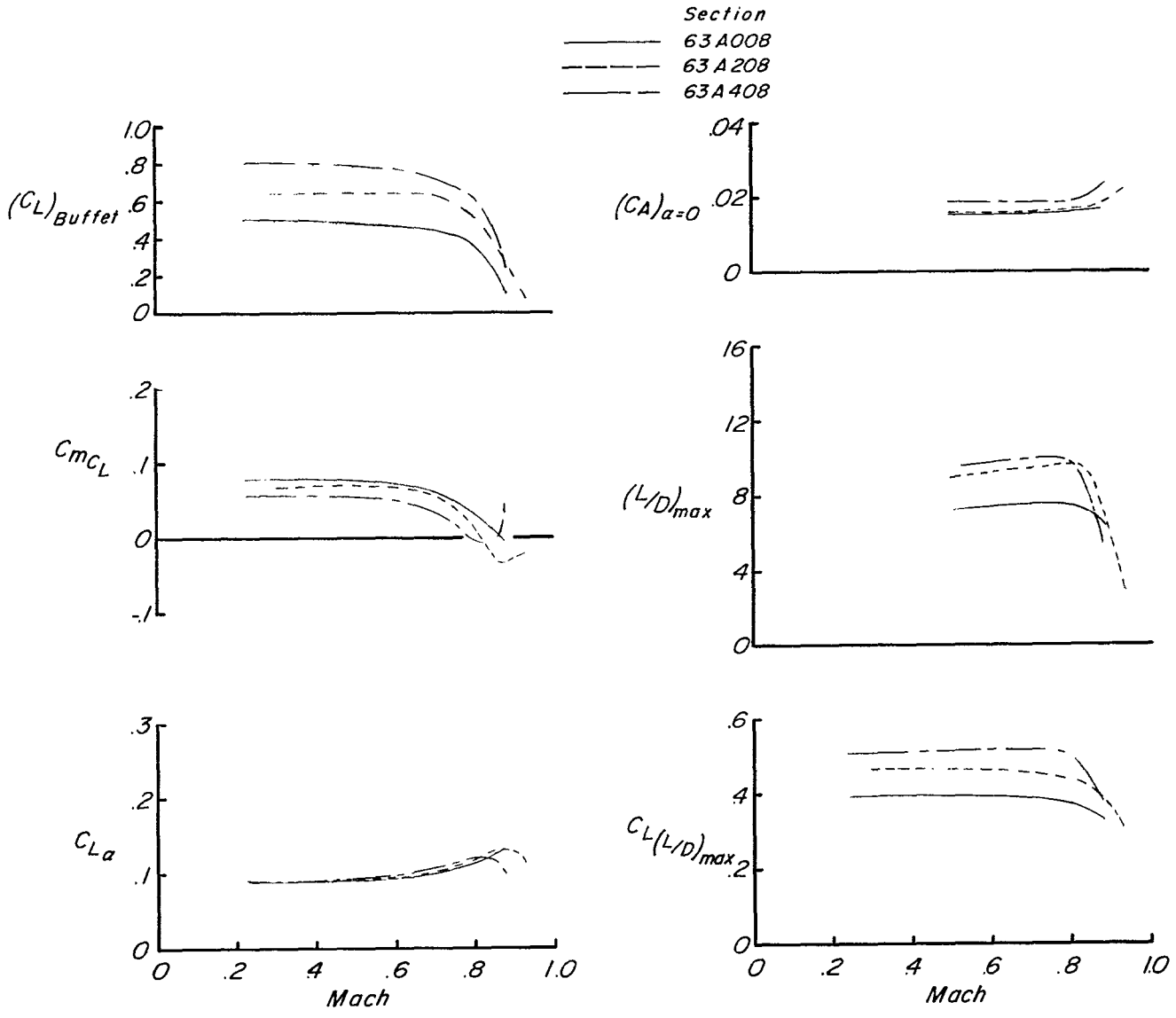
(b) Effects of thickness-to-chord ratio. $\Lambda_{c/4} = 35^0$; A = 6.

Figure 25.- Continued.



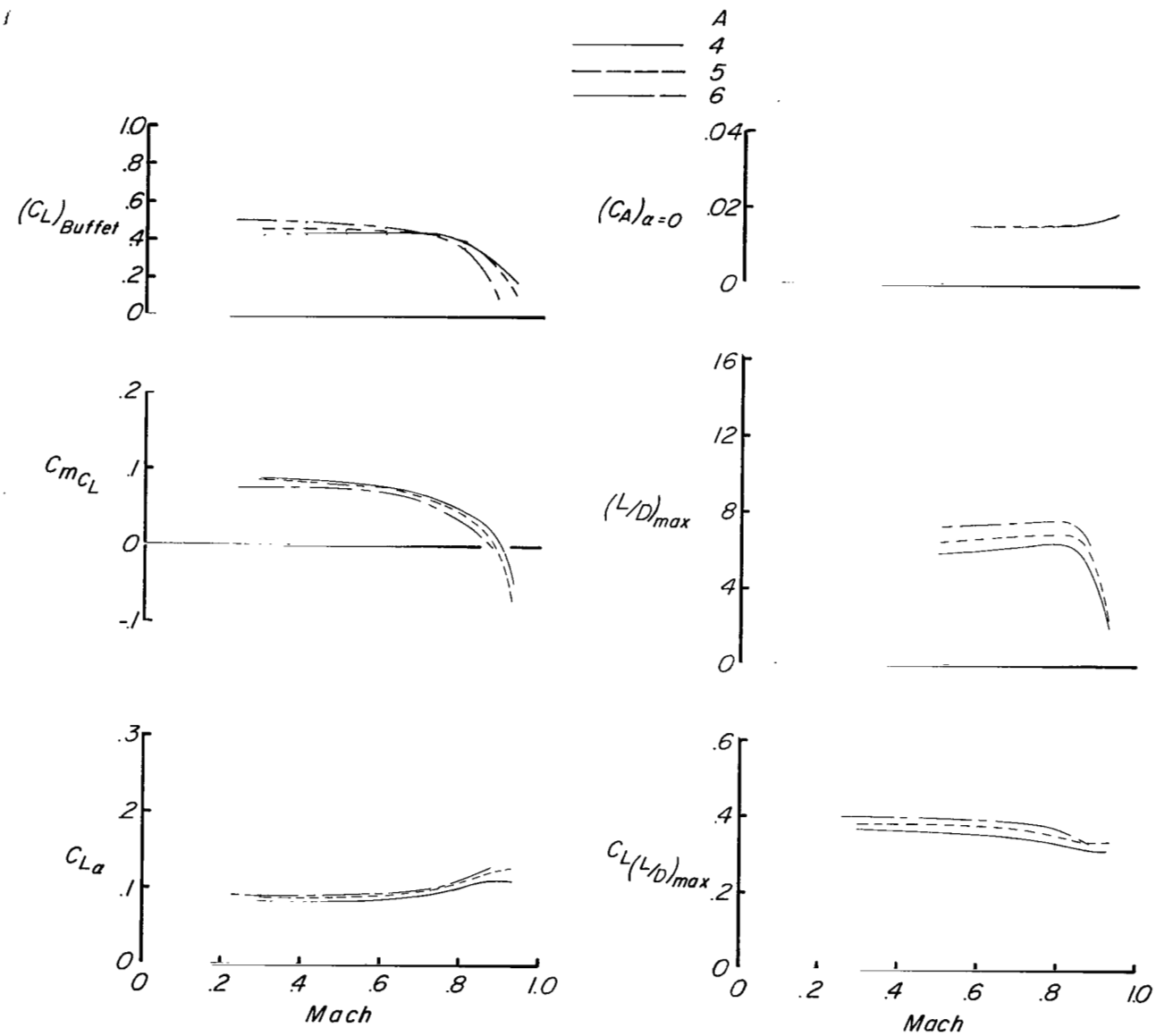
(c) Effects of position-of-maximum thickness. $\Lambda_{c/4} = 35^0$; A = 6.

Figure 25.- Continued.



(d) Effects of camber. $\Lambda_{c/4} = 35^{\circ}$; A = 6.

Figure 25.- Continued.



(e) Effects of aspect ratio. Airfoil, NACA 63A008; $\Lambda_{c/4} = 35^0$.

Figure 25.- Concluded.

"The aeronautical and space activities of the United States shall be conducted so as to contribute . . . to the expansion of human knowledge of phenomena in the atmosphere and space. The Administration shall provide for the widest practicable and appropriate dissemination of information concerning its activities and the results thereof."

—NATIONAL AERONAUTICS AND SPACE ACT OF 1958

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