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PRESSURE DISTRIBUTIONS ON A WING HAVING NACA 4415 AIRFOIL SECTIONS WITH TRAILING-EDGE FLAPS SET AT 0° AND 40°

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16. Abstract

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The results of the investigation are presented as curves of chordwise pressure distributions at the 25-percent-semispan wing station. Tabulated pressure data are also presented as well as the longitudinal aerodynamic characteristics of the model corresponding to the pressure data.

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PRESSURE DISTRIBUTIONS ON A WING HAVING NACA 4415 AIRFOIL SECTIONS WITH TRAILING-EDGE FLAPS SET AT 0° AND 40°

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SUMMARY

An investigation has been made in the Langley 300-MPH 7- by 10-foot tunnel through a range of free-stream dynamic pressures and through a range of angles of attack to determine the chordwise pressure distributions on a wing having NACA 4415 airfoil sections with flaps set at 0° and deflected down 40°. The unswept, untapered wing had an aspect ratio of 6 and full-span 30-percent-chord slotted flaps.

The results of the investigation are presented as curves of chordwise pressure distributions at the 25-percent-semispan wing station. Tabulated pressure data are also presented as well as the longitudinal aerodynamic characteristics of the model corresponding to the pressure data.

INTRODUCTION

A wind-tunnel investigation of the jet-location interference effects on the longitudinal aerodynamic characteristics of a jet V/STOL model has been made for an unswept, untapered wing with an aspect ratio of 6 and 30-percent-chord slotted flaps. Aerodynamic force and pitching-moment measurements for this wing are presented in reference 1. In order to supplement these data, additional aerodynamic and pressure-distribution measurements on the wing are presented in the present paper for the wing in the basic configuration (zero flaps) and with the Fowler-type full-span flaps deflected down 40°. Pressure distributions on the NACA 4415 airfoil section have been presented in reference 2 at Mach numbers from 0.3 to approximately 0.85.

SYMBOLS

The physical quantities in this paper are given both in the International System of Units (SI) and in the U.S. Customary Units. The measurements and calculations were made in the U.S. Customary Units.

C_D drag coefficient, $\frac{\text{Drag}}{q_\infty S}$

C_L lift coefficient, $\frac{\text{Lift}}{q_\infty S}$

C_m pitching-moment coefficient, $\frac{\text{Pitching moment}}{q_\infty Sc}$

C_p' pressure coefficient, $\frac{p_{t,\infty} - p_l}{q_\infty}$

c chord, meters (feet)

p_l local static pressure, newtons/meter^2 (pounds force/foot^2)

$p_{t,\infty}$ free-stream total pressure, newtons/meter^2 (pounds force/foot^2)

q_∞ free-stream dynamic pressure, newtons/meter^2 (pounds force/foot^2)

S wing area, meters^2 (feet^2)

x longitudinal distance from leading edge of wing or flap (positive when measured aft of leading edge of wing), meters (feet)

α angle of attack, degrees

δ_f flap deflection (positive when deflected down), degrees

MODEL AND APPARATUS

A drawing of the model is shown in figure 1. The wing was unswept, untapered, and had an aspect ratio of 6 with a 30-percent wing-chord slotted Fowler-type flap deflected 40°. The flap was removed from the wing for the zero-flap or basic configuration. The wing and flap both were constructed with NACA 4415 airfoil-section contours. The wing was mounted with the top of the wing at the top of a cylindrical fuselage which had a faired nose section. The wing was set at zero incidence relative to the fuselage. In order to determine the chordwise pressure distributions on the wing and flap, pressure orifices were located on the upper and lower surfaces of the right wing and flap at the 25-percent-semispan station. Measurements of the pressures were obtained by the use of pressure transducers.

The model was mounted on a sting-supported six-component strain-gage balance for direct measurement of the total forces and moments on the model as shown in figure 2. The balance was located at the center of the fuselage with the moment center of the balance located at the 25-percent-chord station of the wing. An electronic clinometer was located in the nose section of the fuselage for use in determining the geometric angle of attack of the wing during the investigation.

TEST CONDITIONS

The investigation was made in the Langley 300-MPH 7- by 10-foot tunnel. The Reynolds number based on the maximum free-stream dynamic pressure of 2873 N/m^2 (60 lbf/ft^2) and wing chord of 0.2032 m (0.667 ft) was 0.9×10^6.

Aerodynamic and pressure data were obtained for the wing with flaps deflected down 40° and for the basic wing (flaps zero). The data were obtained at a constant geometric angle of attack of approximately 0° for a range of values of free-stream dynamic pressure from 0 to approximately 2873 N/m^2 (60 lbf/ft^2). Data were also obtained through a range of values of angle of attack at constant free-stream dynamic pressures of approximately 420, 1020, and 2840 N/m^2 (8.8, 21.3, and 59.3 lbf/ft^2).

PRESENTATION OF DATA

The nondimensional pressure coefficients were normalized to the tunnel free-stream dynamic pressure. No corrections were made to the data for model blockage or wind-tunnel boundaries. No corrections or adjustments were made to the longitudinal drag coefficient C_D for the base drag.

The data are presented as indicated below without analysis or discussion:

Type of data	Angle of attack of wing, deg	Flap deflection, deg	Free-stream dynamic pressure		Table	Figure
			N/m^2	lbf/ft^2		
Pressure distribution tabulation	≈0	0	185 to 2847	3.863 to 59.458	I	
	-4.94 to 20.34	0	420	8.8	II	
	-5.01 to 20.59	0	1020	21.3	III	
	-5.10 to 21.22	0	2840	59.3	IV	
	≈0	40	217 to 2320	4.535 to 48.461	V	
	-4.80 to 14.33	40	420	8.8	VI	
	-4.61 to 12.73	40	1020	21.3	VII	
		40	[a]2840	59.3		
Static longitudinal stability	≈0	0, 40	0 to 2873	0 to 60		3
	-5 to 21	0, 40	420	8.8		4
			1020	21.3		
			2840	59.3		
Pressure distribution plots	≈0	0	185 to 2847	3.863 to 59.458		5
	-4.94 to 2.02	0	420	8.8		6(a)
	3.07 to 10.23	0	420	8.8		6(b)
	11.28 to 16.38	0	420	8.8		6(c)
	17.36 to 20.34	0	420	8.8		6(d)
	-5.01 to 2.11	0	1020	21.3		7(a)
	3.13 to 10.40	0	1020	21.3		7(b)
	11.46 to 16.58	0	1020	21.3		7(c)
	17.58 to 20.59	0	1020	21.3		7(d)
	-5.10 to 2.34	0	2840	59.3		8(a)
	3.43 to 10.95	0	2840	59.3		8(b)
	12.00 to 17.15	0	2840	59.3		8(c)
	18.15 to 21.22	0	2840	59.3		8(d)
	≈0	40	217 to 2320	4.535 to 48.461		9
	-4.80 to 2.17	40	420	8.8		10(a)
	3.18 to 8.28	40	420	8.8		10(b)
	9.33 to 14.33	40	420	8.8		10(c)
	-4.61 to 2.55	40	1020	21.3		11(a)
	3.56 to 8.75	40	1020	21.3		11(b)
	9.74 to 12.73	40	1020	21.3		11(c)
		40	[a]2840	59.3		

[a]During the investigation of the wing with flaps deflected down 40° at a free-stream dynamic pressure of 2840 N/m^2 (59.3 lbf/ft^2), the pressures over a large portion of the chords of the wing and flap exceeded the limits of the data-acquisition system, and these pressure data, therefore, are not available.

Langley Research Center,
National Aeronautics and Space Administration,
Hampton, Va., March 24, 1971.

REFERENCES

1. Carter, Arthur W.: Effects of Jet-Exhaust Location on the Longitudinal Aerodynamic Characteristics of a Jet V/STOL Model. NASA TN D-5333, 1969.

2. Graham, Donald J.; Nitzberg, Gerald E.; and Olson, Robert N.: A Systematic Investigation of Pressure Distributions at High Speeds Over Five Representative NACA Low-Drag and Conventional Airfoil Sections. NACA Rep. 832, 1945.

TABLE I

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = .010 DEG DYNAMIC PRESSURE = 3.863 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	CP	X/C	CP
0.0000	.0326	0.0000	.0326
.0050	.2442	.0050	.5861
.0100	.6350	.0100	.8466
.0175	.8141	.0175	1.0583
.0500	1.5142	.0500	1.4653
.0750	1.3677	.0750	1.4328
.1000	1.5305	.1000	1.4165
.2000	1.8561	.2000	1.2801
.3000	1.9050	.3000	1.1648
.4000	1.8235	.4000	1.1302
.5000	1.7421	.5000	1.0764
.6000	1.6282	.6000	1.0648
.7000	1.5793	.7000	1.0341
.8000	1.2374	.8000	1.0072
.9000	1.0420	.9000	.9265
1.0000	1.4165	1.0000	1.4165

ALPHA = .020 DEG DYNAMIC PRESSURE = 8.876 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	CP	X/C	CP
0.0000	.0425	0.0000	.0425
.0050	.1063	.0050	.7086
.0100	.3189	.0100	1.1479
.0175	.5952	.0175	1.3676
.0500	1.3109	.0500	1.5589
.0750	1.4668	.0750	1.5589
.1000	1.6298	.1000	1.5306
.2000	1.7715	.2000	1.3033
.3000	1.7786	.3000	1.1795
.4000	1.6581	.4000	1.1109
.5000	1.5731	.5000	1.0239
.6000	1.5093	.6000	.9871
.7000	1.3818	.7000	.9436
.8000	1.2259	.8000	.9101
.9000	1.0133	.9000	.8700
1.0000	.8149	1.0000	.8149

ALPHA = .050 DEG DYNAMIC PRESSURE = 15.501 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	CP	X/C	CP
0.0000	0.0000	0.0000	0.0000
.0050	.0487	.0050	.8277
.0100	.3611	.0100	1.1442
.0175	.6330	.0175	1.3674
.0500	1.4404	.0500	1.6798
.0750	1.5053	.0750	1.6392
.1000	1.5987	.1000	1.5419
.2000	1.7650	.2000	1.2857
.3000	1.8218	.3000	1.2148
.4000	1.7204	.4000	1.1218
.5000	1.6190	.5000	1.0347
.6000	1.5540	.6000	.9963
.7000	1.3917	.7000	.9647
.8000	1.2538	.8000	.9360
.9000	.9860	.9000	.9005
1.0000	.8561	1.0000	.8561

Note: The conversion factor from pounds force/square foot to newtons/square meter is 47.880258.

TABLE I. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = .070 DEG DYNAMIC PRESSURE = 21.603 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P	X/C	C_P
0.0000	.0466	0.0000	.0466
.0050	.0961	.0050	.8094
.0100	.4018	.0100	.9841
.0175	.5328	.0175	1.3043
.0500	1.3684	.0500	1.5110
.0750	1.4353	.0750	1.5518
.1000	1.6508	.1000	1.5518
.2000	1.7993	.2000	1.2690
.3000	1.8284	.3000	1.1789
.4000	1.7148	.4000	1.1054
.5000	1.6362	.5000	1.0511
.6000	1.5576	.6000	1.0043
.7000	1.3888	.7000	.9754
.8000	1.2519	.8000	.9356
.9000	1.0248	.9000	.8943
1.0000	.8472	1.0000	.8472

ALPHA = .100 DEG DYNAMIC PRESSURE = 31.082 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P	X/C	C_P
0.0000	.0445	0.0000	.0445
.0050	.0668	.0050	.7366
.0100	.4715	.0100	1.0563
.0175	.6637	.0175	1.1818
.0500	1.3699	.0500	1.6735
.0750	1.5703	.0750	1.5986
.1000	1.6755	.1000	1.5116
.2000	1.7483	.2000	1.2986
.3000	1.8333	.3000	1.1983
.4000	1.7322	.4000	1.0989
.5000	1.6310	.5000	1.0272
.6000	1.5035	.6000	1.0081
.7000	1.3841	.7000	.9555
.8000	1.2465	.8000	.9307
.9000	1.0381	.9000	.8887
1.0000	.8560	1.0000	.8560

ALPHA = .130 DEG DYNAMIC PRESSURE = 40.636 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P	X/C	C_P
0.0000	.0201	0.0000	.0201
.0050	.1238	.0050	.9055
.0100	.3467	.0100	1.0525
.0175	.5588	.0175	1.2460
.0500	1.3837	.0500	1.5741
.0750	1.4503	.0750	1.5989
.1000	1.6979	.1000	1.5586
.2000	1.7923	.2000	1.2577
.3000	1.8078	.3000	1.1503
.4000	1.7475	.4000	1.1119
.5000	1.6654	.5000	1.0441
.6000	1.5029	.6000	1.0028
.7000	1.3946	.7000	.9699
.8000	1.2553	.8000	.9348
.9000	1.0194	.9000	.8968
1.0000	.9055	1.0000	.9055

TABLE I. - CONCLUDED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = .160 DEG DYNAMIC PRESSURE = 48.733 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'P	X/C	C'P
0.0000	.0426	0.0000	.0426
.0050	.1071	.0050	.6130
.0100	.3691	.0100	1.0273
.0175	.5782	.0175	1.3087
.0500	1.3332	.0500	1.5229
.0750	1.5229	.0750	1.5978
.1000	1.5655	.1000	1.5436
.2000	1.7346	.2000	1.2669
.3000	1.7707	.3000	1.1584
.4000	1.6894	.4000	1.1085
.5000	1.6184	.5000	1.0329
.6000	1.4868	.6000	1.0049
.7000	1.3861	.7000	.9473
.8000	1.2325	.8000	.9302
.9000	1.0054	.9000	.8852
1.0000	.8337	1.0000	.8337

ALPHA = .200 DEG DYNAMIC PRESSURE = 59.458 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'P	X/C	C'P
0.0000	.0772	0.0000	.0772
.0050	.0878	.0050	.7616
.0100	.3512	.0100	1.1181
.0175	.6220	.0175	1.2747
.0500	1.3815	.0500	1.6248
.0750	1.5783	.0750	1.6005
.1000	1.6851	.1000	1.5053
.2000	1.8131	.2000	1.3081
.3000	1.8163	.3000	1.1905
.4000	1.7264	.4000	1.1295
.5000	1.6513	.5000	1.0439
.6000	1.5032	.6000	1.0171
.7000	1.3942	.7000	.9687
.8000	1.2451	.8000	.9457
.9000	1.0271	.9000	.9085
1.0000	.8568	1.0000	.8568

TABLE II

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -4.940 DEG DYNAMIC PRESSURE = 8.821 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	1.1439	0.0000	1.1439
.0050	.0957	.0050	2.5200
.0100	.0458	.0100	2.8623
.0175	.1100	.0175	3.0120
.0500	.6947	.0500	2.6270
.0750	.8872	.0750	2.5200
.1000	.8872	.1000	2.2063
.2000	1.3792	.2000	1.6650
.3000	1.5717	.3000	1.4444
.4000	1.5361	.4000	1.3182
.5000	1.5075	.5000	1.1953
.6000	1.4434	.6000	1.1296
.7000	1.4077	.7000	1.0724
.8000	1.3221	.8000	1.0168
.9000	1.0227	.9000	.9663
1.0000	1.2009	1.0000	1.2009

ALPHA = -4.040 DEG DYNAMIC PRESSURE = 8.766 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.6601	0.0000	.6601
.0050	.0430	.0050	2.0305
.0100	.0646	.0100	2.2816
.0175	.1004	.0175	2.4825
.0500	.8323	.0500	2.4969
.0750	1.0619	.0750	2.3247
.1000	1.1910	.1000	2.1453
.2000	1.4852	.2000	1.5772
.3000	1.6072	.3000	1.3756
.4000	1.5570	.4000	1.2892
.5000	1.5283	.5000	1.1892
.6000	1.4709	.6000	1.1198
.7000	1.3919	.7000	1.0435
.8000	1.3417	.8000	.9961
.9000	1.0188	.9000	.9368
1.0000	1.1552	1.0000	1.1552

ALPHA = -2.990 DEG DYNAMIC PRESSURE = 8.803 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.5073	0.0000	.5073
.0050	.0214	.0050	1.6219
.0100	.0429	.0100	2.2721
.0175	.1572	.0175	2.5150
.0500	.7931	.0500	2.2649
.0750	1.0717	.0750	2.0077
.1000	1.2575	.1000	1.7934
.2000	1.5147	.2000	1.4980
.3000	1.5790	.3000	1.3361
.4000	1.5576	.4000	1.2332
.5000	1.5361	.5000	1.1302
.6000	1.4647	.6000	1.0881
.7000	1.4004	.7000	1.0172
.8000	1.3432	.8000	.9632
.9000	1.0074	.9000	.9093
1.0000	1.1503	1.0000	1.1503

Note: The conversion factor from pounds force/square foot to newtons/square meter is 47.880258.

TABLE II. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -1.970 DEG DYNAMIC PRESSURE = 8.858 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'P	X/C	C'P
0.0000	.0071	0.0000	.0071
.0050	-.1491	.0050	.9373
.0100	0.0000	.0100	1.0651
.0175	.1349	.0175	1.6189
.0500	.9231	.0500	1.7822
.0750	1.1929	.0750	1.6970
.1000	1.1858	.1000	1.6260
.2000	1.4201	.2000	1.3613
.3000	1.4982	.3000	1.2171
.4000	1.4343	.4000	1.1651
.5000	1.4201	.5000	1.0796
.6000	1.3349	.6000	1.0076
.7000	1.2568	.7000	.9606
.8000	1.0793	.8000	.9422
.9000	.8947	.9000	.8952
1.0000	.9870	1.0000	.9870

ALPHA = -1.000 DEG DYNAMIC PRESSURE = 8.849 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'P	X/C	C'P
0.0000	.0995	0.0000	.0995
.0050	.0711	.0050	1.0448
.0100	.2843	.0100	1.1657
.0175	.5118	.0175	1.6703
.0500	1.1657	.0500	1.9973
.0750	1.2083	.0750	1.8054
.1000	1.3931	.1000	1.6632
.2000	1.6348	.2000	1.3409
.3000	1.7130	.3000	1.1949
.4000	1.6774	.4000	1.1361
.5000	1.5992	.5000	1.0640
.6000	1.5210	.6000	1.0304
.7000	1.4286	.7000	.9935
.8000	1.1728	.8000	.9549
.9000	1.0235	.9000	.9062
1.0000	1.1443	1.0000	1.1443

ALPHA = .020 DEG DYNAMIC PRESSURE = 8.840 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'P	X/C	C'P
0.0000	.0569	0.0000	.0569
.0050	.1636	.0050	.6261
.0100	.4127	.0100	.9747
.0175	.6617	.0175	1.3163
.0500	1.4159	.0500	1.5937
.0750	1.5297	.0750	1.5297
.1000	1.5511	.1000	1.5511
.2000	1.6862	.2000	1.2969
.3000	1.7858	.3000	1.1625
.4000	1.7289	.4000	1.1087
.5000	1.6293	.5000	1.0398
.6000	1.5155	.6000	1.0029
.7000	1.4301	.7000	.9693
.8000	1.2807	.8000	.9494
.9000	1.0459	.9000	.9155
1.0000	1.1740	1.0000	1.1740

TABLE II. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 1.030 DEG DYNAMIC PRESSURE = 8.849 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	CP	X/C	CP
0.0000	.0213	0.0000	.0213
.0050	.2203	.0050	.3767
.0100	.5970	.0100	.6326
.0175	.7818	.0175	.9951
.0500	1.4073	.0500	1.4642
.0750	1.6134	.0750	1.4855
.1000	1.7982	.1000	1.4073
.2000	1.9120	.2000	1.2368
.3000	1.9475	.3000	1.1495
.4000	1.8196	.4000	1.0891
.5000	1.6916	.5000	1.0287
.6000	1.6206	.6000	1.0002
.7000	1.4358	.7000	.9599
.8000	1.2865	.8000	.9381
.9000	1.0733	.9000	.8928
1.0000	1.1657	1.0000	1.1657

ALPHA = 2.020 DEG DYNAMIC PRESSURE = 8.821 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	CP	X/C	CP
0.0000	.0143	0.0000	.0143
.0050	.2709	.0050	.2995
.0100	.7344	.0100	.6203
.0175	1.0196	.0175	.9483
.0500	1.6970	.0500	1.3334
.0750	1.7683	.0750	1.3262
.1000	1.9323	.1000	1.3048
.2000	1.9965	.2000	1.1869
.3000	1.9893	.3000	1.1246
.4000	1.8111	.4000	1.0690
.5000	1.7184	.5000	.9949
.6000	1.6400	.6000	.9798
.7000	1.4474	.7000	.9495
.8000	1.2977	.8000	.9259
.9000	1.0481	.9000	.8855
1.0000	1.1480	1.0000	1.1480

ALPHA = 3.070 DEG DYNAMIC PRESSURE = 8.803 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	CP	X/C	CP
0.0000	.0357	0.0000	.0357
.0050	.6645	.0050	.1666
.0100	1.1074	.0100	.4215
.0175	1.1789	.0175	.6645
.0500	1.9005	.0500	1.0717
.0750	1.9362	.0750	1.1146
.1000	1.9577	.1000	1.1575
.2000	2.0220	.2000	1.0780
.3000	2.0577	.3000	1.0240
.4000	1.8648	.4000	.9987
.5000	1.7934	.5000	.9717
.6000	1.5790	.6000	.9548
.7000	1.4718	.7000	.9278
.8000	1.2932	.8000	.9076
.9000	1.0574	.9000	.8823
1.0000	1.1932	1.0000	1.1932

TABLE II. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 4.100 DEG DYNAMIC PRESSURE = 8.821 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P	X/C	C_P
0.0000	.0784	0.0000	.0784
.0050	.8842	.0050	.0357
.0100	1.3048	.0100	.1925
.0175	1.4902	.0175	.4920
.0500	2.0749	.0500	1.0410
.0750	2.0393	.0750	1.1194
.1000	2.1248	.1000	1.2193
.2000	2.1319	.2000	1.0303
.3000	2.0464	.3000	.9916
.4000	1.9038	.4000	.9764
.5000	1.8325	.5000	.9529
.6000	1.5900	.6000	.9495
.7000	1.4688	.7000	.9175
.8000	1.3048	.8000	.8956
.9000	1.0624	.9000	.8771
1.0000	1.1908	1.0000	1.1908

ALPHA = 5.090 DEG DYNAMIC PRESSURE = 8.794 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P	X/C	C_P
0.0000	.2575	0.0000	.2575
.0050	1.0442	.0050	.0572
.0100	1.5735	.0100	.1811
.0175	1.9597	.0175	.5293
.0500	2.3888	.0500	1.0013
.0750	2.3960	.0750	1.1229
.1000	2.3030	.1000	1.1014
.2000	2.1528	.2000	.9321
.3000	2.0956	.3000	.9642
.4000	2.0098	.4000	.9440
.5000	1.8381	.5000	.8967
.6000	1.6164	.6000	.9119
.7000	1.4948	.7000	.8714
.8000	1.3160	.8000	.8832
.9000	1.0728	.9000	.8714
1.0000	1.1801	1.0000	1.1801

ALPHA = 6.140 DEG DYNAMIC PRESSURE = 8.721 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P	X/C	C_P
0.0000	.2885	0.0000	.2885
.0050	1.1106	.0050	.0216
.0100	1.7237	.0100	.1226
.0175	1.8679	.0175	.3101
.0500	2.7694	.0500	.6491
.0750	2.5170	.0750	.7212
.1000	2.5026	.1000	.8582
.2000	2.3728	.2000	.8940
.3000	2.2285	.3000	.9382
.4000	2.0843	.4000	.9212
.5000	1.7597	.5000	.9008
.6000	1.5939	.6000	.8803
.7000	1.5001	.7000	.8786
.8000	1.2188	.8000	.8684
.9000	1.0602	.9000	.8667
1.0000	1.1828	1.0000	1.1828

TABLE II. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 7.150 DEG DYNAMIC PRESSURE = 8.766 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.5094	0.0000	.5094
.0050	1.6287	.0050	.0430
.0100	1.7292	.0100	.0430
.0175	2.1668	.0175	.1148
.0500	2.7050	.0500	.5596
.0750	2.5902	.0750	.7318
.1000	2.5112	.1000	.7964
.2000	2.4180	.2000	.8301
.3000	2.2888	.3000	.8674
.4000	2.1453	.4000	.9165
.5000	1.7579	.5000	.8792
.6000	1.6359	.6000	.8792
.7000	1.4852	.7000	.8657
.8000	1.2915	.8000	.8792
.9000	1.0188	.9000	.8674
1.0000	1.1982	1.0000	1.1982

ALPHA = 8.190 DEG DYNAMIC PRESSURE = 8.693 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.6078	0.0000	.6078
.0050	2.1489	.0050	.0506
.0100	2.7349	.0100	0.0000
.0175	2.6481	.0175	.0289
.0500	2.9809	.0500	.4341
.0750	2.7783	.0750	.6584
.1000	2.7132	.1000	.7597
.2000	2.5468	.2000	.8063
.3000	2.3153	.3000	.8200
.4000	2.1489	.4000	.8234
.5000	1.8160	.5000	.8371
.6000	1.6207	.6000	.8593
.7000	1.4977	.7000	.8610
.8000	1.2806	.8000	.8644
.9000	1.0419	.9000	.8729
1.0000	1.1721	1.0000	1.1721

ALPHA = 9.230 DEG DYNAMIC PRESSURE = 8.730 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.8573	0.0000	.8573
.0050	2.2262	.0050	.1009
.0100	2.6657	.0100	-.0072
.0175	2.9899	.0175	.0144
.0500	3.1988	.0500	.4179
.0750	2.9611	.0750	.5331
.1000	2.8674	.1000	.6196
.2000	2.5072	.2000	.7485
.3000	2.3775	.3000	.8012
.4000	2.0317	.4000	.8233
.5000	1.8228	.5000	.8131
.6000	1.6715	.6000	.8386
.7000	1.4625	.7000	.8386
.8000	1.1960	.8000	.8624
.9000	1.0230	.9000	.8743
1.0000	1.1671	1.0000	1.1671

TABLE II. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 10.230 DEG DYNAMIC PRESSURE = 8.702 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	1.1998	0.0000	1.1998
.0050	2.9056	.0050	.2313
.0100	3.0863	.0100	.0072
.0175	3.2886	.0175	-.0289
.0500	3.5416	.0500	.2819
.0750	3.1585	.0750	.3469
.1000	2.9995	.1000	.5782
.2000	2.5948	.2000	.6980
.3000	2.4719	.3000	.7355
.4000	2.0310	.4000	.7969
.5000	1.8503	.5000	.8038
.6000	1.6552	.6000	.8242
.7000	1.4672	.7000	.8311
.8000	1.2287	.8000	.8652
.9000	1.0408	.9000	.8772
1.0000	1.1564	1.0000	1.1564

ALPHA = 11.280 DEG DYNAMIC PRESSURE = 8.702 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	1.4094	0.0000	1.4094
.0050	2.9778	.0050	.4481
.0100	3.3826	.0100	.1012
.0175	3.5705	.0175	.0434
.0500	3.6139	.0500	.2602
.0750	3.2308	.0750	.4337
.1000	2.9923	.1000	.4770
.2000	2.7538	.2000	.6707
.3000	2.4791	.3000	.7406
.4000	2.1539	.4000	.7696
.5000	1.9154	.5000	.7799
.6000	1.6841	.6000	.8123
.7000	1.4889	.7000	.8345
.8000	1.2721	.8000	.8601
.9000	1.1058	.9000	.8754
1.0000	1.1926	1.0000	1.1926

ALPHA = 12.270 DEG DYNAMIC PRESSURE = 8.656 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	1.8311	0.0000	1.8311
.0050	3.5023	.0050	.3488
.0100	3.8511	.0100	.0945
.0175	3.8365	.0175	-.0363
.0500	3.9165	.0500	.2616
.0750	3.4296	.0750	.2979
.1000	3.3933	.1000	.4505
.2000	2.7393	.2000	.6416
.3000	2.4342	.3000	.7068
.4000	2.0927	.4000	.7669
.5000	1.8674	.5000	.7823
.6000	1.5913	.6000	.8046
.7000	1.3733	.7000	.8355
.8000	1.2643	.8000	.8818
.9000	1.1190	.9000	.9161
1.0000	1.1408	1.0000	1.1408

TABLE II. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 13.290 DEG DYNAMIC PRESSURE = 8.711 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	1.8917	0.0000	1.8917
.0050	4.1517	.0050	.4260
.0100	3.9928	.0100	.2455
.0175	3.9423	.0175	.0289
.0500	3.8773	.0500	.1516
.0750	3.5307	.0750	.2744
.1000	3.3935	.1000	.4043
.2000	2.8159	.2000	.6137
.3000	2.4549	.3000	.6700
.4000	2.1228	.4000	.7501
.5000	1.8123	.5000	.7859
.6000	1.6173	.6000	.8217
.7000	1.3935	.7000	.8592
.8000	1.2780	.8000	.9052
.9000	1.1769	.9000	.9513
1.0000	1.1119	1.0000	1.1119

ALPHA = 14.300 DEG DYNAMIC PRESSURE = 8.721 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	2.3872	0.0000	2.3872
.0050	3.9233	.0050	1.0313
.0100	3.9955	.0100	.3029
.0175	4.2190	.0175	.0938
.0500	4.1325	.0500	.0865
.0750	3.6998	.0750	.2596
.1000	3.4546	.1000	.3534
.2000	2.8343	.2000	.5585
.3000	2.4160	.3000	.6743
.4000	2.0122	.4000	.7373
.5000	1.6804	.5000	.7526
.6000	1.5145	.6000	.7986
.7000	1.4063	.7000	.8361
.8000	1.2405	.8000	.8906
.9000	1.1323	.9000	.9672
1.0000	1.1900	1.0000	1.1900

ALPHA = 15.350 DEG DYNAMIC PRESSURE = 8.657 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	3.0369	0.0000	3.0369
.0050	4.7951	.0050	1.2133
.0100	4.8242	.0100	.4940
.0175	4.7879	.0175	.1380
.0500	4.3665	.0500	.1235
.0750	3.8942	.0750	.2252
.1000	3.4874	.1000	.3705
.2000	2.8262	.2000	.4975
.3000	2.4194	.3000	.6193
.4000	1.8599	.4000	.7033
.5000	1.4167	.5000	.7393
.6000	1.4095	.6000	.7977
.7000	1.3877	.7000	.8371
.8000	1.3223	.8000	.8989
.9000	1.3150	.9000	.9778
1.0000	1.2278	1.0000	1.2278

TABLE II. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 16.380 DEG DYNAMIC PRESSURE = 8.745 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	3.0351	0.0000	3.0351
.0050	4.7397	.0050	1.1436
.0100	4.7181	.0100	.5106
.0175	4.6749	.0175	.1151
.0500	4.3009	.0500	.0360
.0750	3.6968	.0750	.1438
.1000	3.4235	.1000	.3021
.2000	2.6036	.2000	.4891
.3000	1.9779	.3000	.6147
.4000	1.4528	.4000	.6962
.5000	1.4025	.5000	.7404
.6000	1.3953	.6000	.8032
.7000	1.3953	.7000	.8440
.8000	1.5104	.8000	.9034
.9000	1.4313	.9000	1.0274
1.0000	1.1723	1.0000	1.1723

ALPHA = 17.360 DEG DYNAMIC PRESSURE = 9.069 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	3.2318	0.0000	3.2318
.0050	5.0697	.0050	1.3801
.0100	4.9657	.0100	.6589
.0175	4.9310	.0175	.2150
.0500	4.4732	.0500	.0763
.0750	3.7936	.0750	.1664
.1000	3.4468	.1000	.2774
.2000	2.3857	.2000	.4945
.3000	1.5674	.3000	.6337
.4000	1.5050	.4000	.7156
.5000	1.4495	.5000	.7696
.6000	1.5535	.6000	.8400
.7000	1.5466	.7000	.8842
.8000	1.5604	.8000	.9743
.9000	1.5327	.9000	1.1004
1.0000	1.1929	1.0000	1.1929

ALPHA = 18.360 DEG DYNAMIC PRESSURE = 8.684 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	3.2592	0.0000	3.2592
.0050	5.1061	.0050	1.4051
.0100	5.0337	.0100	.6663
.0175	4.9613	.0175	.1521
.0500	4.5702	.0500	.0290
.0750	3.8749	.0750	.1159
.1000	3.6518	.1000	.2390
.2000	2.7233	.2000	.5062
.3000	2.2742	.3000	.6276
.4000	1.8976	.4000	.7199
.5000	1.5644	.5000	.7661
.6000	1.4268	.6000	.8294
.7000	1.4413	.7000	.8773
.8000	1.3616	.8000	.9611
.9000	1.4558	.9000	1.0722
1.0000	1.2384	1.0000	1.2385

TABLE II. - CONCLUDED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 19.340 DEG DYNAMIC PRESSURE = 8.611 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	3.5060	0.0000	3.5060
.0050	5.1640	.0050	1.6215
.0100	5.2517	.0100	.7596
.0175	5.0837	.0175	.2337
.0500	4.5797	.0500	.0511
.0750	3.8566	.0750	.1680
.1000	3.3380	.1000	.2483
.2000	2.1766	.2000	.4674
.3000	2.3227	.3000	.6070
.4000	1.9502	.4000	.6915
.5000	1.5120	.5000	.7433
.6000	1.4900	.6000	.7967
.7000	1.4097	.7000	.8519
.8000	1.3367	.8000	.9502
.9000	1.4754	.9000	1.0830
1.0000	1.3147	1.0000	1.3147

ALPHA = 20.340 DEG DYNAMIC PRESSURE = 8.757 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	3.4116	0.0000	3.4116
.0050	5.1498	.0050	1.6232
.0100	5.3078	.0100	.7901
.0175	5.1785	.0175	.2801
.0500	4.5680	.0500	0.0000
.0750	3.8569	.0750	.1006
.1000	3.4260	.1000	.2083
.2000	2.3271	.2000	.4528
.3000	1.9895	.3000	.5918
.4000	1.6376	.4000	.6664
.5000	1.5227	.5000	.7122
.6000	1.4365	.6000	.7852
.7000	1.4580	.7000	.8513
.8000	1.4939	.8000	.9378
.9000	1.5298	.9000	1.0650
1.0000	1.2928	1.0000	1.2928

TABLE III

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -5.010 DEG DYNAMIC PRESSURE = 21.238 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.7493	0.0000	.7493
.0050	.0385	.0050	2.3218
.0100	.0059	.0100	2.6564
.0175	.0296	.0175	2.9437
.0500	.5597	.0500	2.5025
.0750	.8736	.0750	2.2537
.1000	1.0632	.1000	2.1056
.2000	1.3564	.2000	1.5684
.3000	1.5104	.3000	1.3929
.4000	1.4689	.4000	1.2684
.5000	1.4334	.5000	1.1838
.6000	1.3949	.6000	1.0929
.7000	1.3416	.7000	1.0132
.8000	1.4156	.8000	.9670
.9000	.9980	.9000	.9055
1.0000	.8884	1.0000	.8884

ALPHA = -4.070 DEG DYNAMIC PRESSURE = 21.384 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.5677	0.0000	.5677
.0050	-.0088	.0050	2.2207
.0100	.0176	.0100	2.5118
.0175	.1324	.0175	2.6618
.0500	.7206	.0500	2.3677
.0750	.9265	.0750	2.1824
.1000	1.1765	.1000	2.0001
.2000	1.5000	.2000	1.5271
.3000	1.5853	.3000	1.3799
.4000	1.5206	.4000	1.2722
.5000	1.5059	.5000	1.1556
.6000	1.4265	.6000	1.1000
.7000	1.3706	.7000	1.0313
.8000	1.2500	.8000	.9743
.9000	1.0000	.9000	.9167
1.0000	.8765	1.0000	.8765

ALPHA = -2.990 DEG DYNAMIC PRESSURE = 21.466 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.4454	0.0000	.4454
.0050	0.0000	.0050	1.5676
.0100	.0264	.0100	1.8928
.0175	.2403	.0175	2.2092
.0500	.8380	.0500	2.1418
.0750	1.0812	.0750	1.9748
.1000	1.2687	.1000	1.8020
.2000	1.5793	.2000	1.4867
.3000	1.6467	.3000	1.3352
.4000	1.5090	.4000	1.2473
.5000	1.4943	.5000	1.1152
.6000	1.4445	.6000	1.0702
.7000	1.3859	.7000	1.0114
.8000	1.2160	.8000	.9616
.9000	1.0079	.9000	.9035
1.0000	.8644	1.0000	.8644

Note: The conversion factor from pounds force/square foot to newtons/square meter is 47.880258.

TABLE III. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -1.990 DEG DYNAMIC PRESSURE = 21.393 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.2146	0.0000	.2146
.0050	0.0000	.0050	1.3612
.0100	.1558	.0100	1.3642
.0175	.3528	.0175	1.7846
.0500	1.0466	.0500	2.0198
.0750	1.2201	.0750	1.7758
.1000	1.4083	.1000	1.7082
.2000	1.5817	.2000	1.3862
.3000	1.6670	.3000	1.2523
.4000	1.5906	.4000	1.1655
.5000	1.5700	.5000	1.0753
.6000	1.4788	.6000	1.0329
.7000	1.4083	.7000	.9743
.8000	1.2231	.8000	.9406
.9000	1.0172	.9000	.8885
1.0000	.8644	1.0000	.8644

ALPHA = -.980 DEG DYNAMIC PRESSURE = 21.466 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.1026	0.0000	.1026
.0050	.0469	.0050	.9669
.0100	.2198	.0100	1.1837
.0175	.4542	.0175	1.6115
.0500	1.0841	.0500	1.8166
.0750	1.1808	.0750	1.7404
.1000	1.4357	.1000	1.6262
.2000	1.6701	.2000	1.3338
.3000	1.7287	.3000	1.2030
.4000	1.6584	.4000	1.1525
.5000	1.6027	.5000	1.0647
.6000	1.5119	.6000	1.0453
.7000	1.3683	.7000	.9941
.8000	1.2248	.8000	.9547
.9000	1.0196	.9000	.9049
1.0000	.8556	1.0000	.8556

ALPHA = .060 DEG DYNAMIC PRESSURE = 21.393 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.1000	0.0000	.1000
.0050	.0676	.0050	.6439
.0100	.3616	.0100	1.0290
.0175	.5557	.0175	1.3877
.0500	1.3024	.0500	1.6229
.0750	1.4083	.0750	1.5817
.1000	1.6082	.1000	1.5641
.2000	1.8434	.2000	1.2988
.3000	1.8904	.3000	1.1912
.4000	1.7346	.4000	1.1266
.5000	1.6376	.5000	1.0565
.6000	1.5494	.6000	1.0183
.7000	1.4142	.7000	.9746
.8000	1.2613	.8000	.9378
.9000	1.0319	.9000	.8962
1.0000	.8673	1.0000	.8673

TABLE III. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 1.070 DEG DYNAMIC PRESSURE = 21.357 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'P	X/C	C'P
0.0000	.0118	0.0000	.0118
.0050	.1414	.0050	.4800
.0100	.4977	.0100	.8452
.0175	.6832	.0175	1.0808
.0500	1.5726	.0500	1.3812
.0750	1.4607	.0750	1.4195
.1000	1.6963	.1000	1.4342
.2000	1.8877	.2000	1.2286
.3000	1.8907	.3000	1.1438
.4000	1.7788	.4000	1.0819
.5000	1.6786	.5000	1.0166
.6000	1.5432	.6000	.9707
.7000	1.3959	.7000	.9540
.8000	1.2516	.8000	.9178
.9000	1.0131	.9000	.8803
1.0000	.8511	1.0000	.8511

ALPHA = 2.110 DEG DYNAMIC PRESSURE = 21.411 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'P	X/C	C'P
0.0000	.0617	0.0000	.0617
.0050	.2967	.0050	.3672
.0100	.7491	.0100	.6316
.0175	.9841	.0175	.8020
.0500	1.7978	.0500	1.2220
.0750	1.8565	.0750	1.3983
.1000	1.9329	.1000	1.3072
.2000	1.9828	.2000	1.1680
.3000	1.9593	.3000	1.0917
.4000	1.7978	.4000	1.0431
.5000	1.7097	.5000	.9863
.6000	1.5217	.6000	.9592
.7000	1.4218	.7000	.9259
.8000	1.2661	.8000	.9065
.9000	1.0193	.9000	.8732
1.0000	.8460	1.0000	.8460

ALPHA = 3.130 DEG DYNAMIC PRESSURE = 21.430 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'P	X/C	C'P
0.0000	.0411	0.0000	.0411
.0050	.5664	.0050	.1908
.0100	.9216	.0100	.4109
.0175	1.1358	.0175	.6369
.0500	1.9400	.0500	1.2239
.0750	1.9224	.0750	1.2297
.1000	2.2012	.1000	1.2121
.2000	2.0691	.2000	1.0685
.3000	2.0427	.3000	1.0401
.4000	1.8901	.4000	1.0193
.5000	1.7492	.5000	.9653
.6000	1.5731	.6000	.9473
.7000	1.4499	.7000	.9216
.8000	1.2708	.8000	.9036
.9000	1.0478	.9000	.8787
1.0000	.8687	1.0000	.8687

TABLE III. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 4.170 DEG DYNAMIC PRESSURE = 21.457 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.0791	0.0000	.0791
.0050	.7621	.0050	.0322
.0100	1.1373	.0100	.1524
.0175	1.4334	.0175	.4925
.0500	2.1369	.0500	.9878
.0750	2.1721	.0750	1.0875
.1000	2.2190	.1000	1.1285
.2000	2.0812	.2000	.9966
.3000	2.0460	.3000	.9973
.4000	1.9141	.4000	.9862
.5000	1.7236	.5000	.9440
.6000	1.5975	.6000	.9309
.7000	1.4715	.7000	.9039
.8000	1.2839	.8000	.8956
.9000	1.0494	.9000	.8727
1.0000	.8559	1.0000	.8559

ALPHA = 5.210 DEG DYNAMIC PRESSURE = 21.329 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.1356	0.0000	.1356
.0050	.9466	.0050	-.0088
.0100	1.3417	.0100	.1180
.0175	1.6690	.0175	.2919
.0500	2.2175	.0500	.7431
.0750	2.4416	.0750	.8699
.1000	2.2087	.1000	1.0793
.2000	2.2205	.2000	.9518
.3000	2.1232	.3000	.9587
.4000	1.9816	.4000	.9552
.5000	1.7251	.5000	.9100
.6000	1.5983	.6000	.9128
.7000	1.4479	.7000	.8877
.8000	1.2680	.8000	.8961
.9000	1.0174	.9000	.8766
1.0000	.8345	1.0000	.8345

ALPHA = 6.260 DEG DYNAMIC PRESSURE = 21.256 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.3551	0.0000	.3551
.0050	1.3907	.0050	.0118
.0100	1.6985	.0100	.0740
.0175	1.9174	.0175	.2249
.0500	2.5299	.0500	.7309
.0750	2.5151	.0750	.9617
.1000	2.3701	.1000	.9291
.2000	2.3701	.2000	.8782
.3000	2.2340	.3000	.8984
.4000	2.0713	.4000	.9103
.5000	1.7695	.5000	.8880
.6000	1.6304	.6000	.8852
.7000	1.4706	.7000	.8712
.8000	1.2783	.8000	.8782
.9000	1.0386	.9000	.8649
1.0000	.8492	1.0000	.8492

TABLE III. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 7.300 DEG DYNAMIC PRESSURE = 21.138 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.5594	0.0000	.5594
.0050	1.8686	.0050	.0119
.0100	2.2048	.0100	.0149
.0175	2.2792	.0175	.0625
.0500	2.6839	.0500	.5118
.0750	2.5738	.0750	.6576
.1000	2.6244	.1000	.8034
.2000	2.4578	.2000	.8107
.3000	2.3239	.3000	.8339
.4000	2.0055	.4000	.8634
.5000	1.8151	.5000	.8557
.6000	1.6365	.6000	.8725
.7000	1.4758	.7000	.8550
.8000	1.2735	.8000	.8606
.9000	1.0265	.9000	.8669
1.0000	.8569	1.0000	.8569

ALPHA = 8.310 DEG DYNAMIC PRESSURE = 21.439 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.5926	0.0000	.5926
.0050	1.9861	.0050	.0147
.0100	2.4056	.0100	0.0000
.0175	2.5025	.0175	.0675
.0500	3.0452	.0500	.4547
.0750	2.9249	.0750	.6161
.1000	2.8662	.1000	.7276
.2000	2.5259	.2000	.7986
.3000	2.3294	.3000	.8277
.4000	2.0272	.4000	.8513
.5000	1.8394	.5000	.8471
.6000	1.6341	.6000	.8534
.7000	1.4727	.7000	.8444
.8000	1.2439	.8000	.8686
.9000	1.0092	.9000	.8707
1.0000	.8742	1.0000	.8742

ALPHA = 9.380 DEG DYNAMIC PRESSURE = 21.512 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.8801	0.0000	.8801
.0050	2.3302	.0050	.1608
.0100	2.5729	.0100	.0497
.0175	2.7659	.0175	.0263
.0500	3.2015	.0500	.3772
.0750	2.9822	.0750	.6023
.1000	2.9121	.1000	.7105
.2000	2.6109	.2000	.7462
.3000	2.3799	.3000	.7918
.4000	2.0817	.4000	.8318
.5000	1.8712	.5000	.8311
.6000	1.6782	.6000	.8332
.7000	1.4911	.7000	.8415
.8000	1.2572	.8000	.8594
.9000	1.0613	.9000	.8733
1.0000	.9531	1.0000	.9531

TABLE III. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 10.400 DEG DYNAMIC PRESSURE = 21.311 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	1.1569	0.0000	1.1569
.0050	2.6887	.0050	.3128
.0100	2.9425	.0100	.1062
.0175	3.3586	.0175	-.0059
.0500	3.6183	.0500	.3483
.0750	3.2612	.0750	.5224
.1000	3.0930	.1000	.5696
.2000	2.6473	.2000	.6989
.3000	2.3758	.3000	.7707
.4000	2.0689	.4000	.8146
.5000	1.8387	.5000	.8062
.6000	1.6321	.6000	.8320
.7000	1.4284	.7000	.8411
.8000	1.2160	.8000	.8717
.9000	1.0389	.9000	.8871
1.0000	.9592	1.0000	.9592

ALPHA = 11.460 DEG DYNAMIC PRESSURE = 21.055 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	1.2546	0.0000	1.2546
.0050	3.0261	.0050	.3525
.0100	3.4383	.0100	.1105
.0175	3.6922	.0175	.0149
.0500	3.8207	.0500	.2569
.0750	3.3547	.0750	.4511
.1000	3.1545	.1000	.5407
.2000	2.6945	.2000	.6877
.3000	2.4077	.3000	.7307
.4000	2.0731	.4000	.7758
.5000	1.8401	.5000	.7794
.6000	1.6041	.6000	.8174
.7000	1.4010	.7000	.8351
.8000	1.1979	.8000	.8710
.9000	1.1531	.9000	.9127
1.0000	1.0635	1.0000	1.0635

ALPHA = 12.500 DEG DYNAMIC PRESSURE = 21.220 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	1.8229	0.0000	1.8229
.0050	3.4827	.0050	.5898
.0100	3.8858	.0100	.1867
.0175	3.6398	.0175	.0326
.0500	3.8354	.0500	.1838
.0750	3.4531	.0750	.3616
.1000	3.2782	.1000	.4416
.2000	2.7565	.2000	.6193
.3000	2.4483	.3000	.7012
.4000	2.0896	.4000	.7537
.5000	1.8258	.5000	.7817
.6000	1.5531	.6000	.8237
.7000	1.3901	.7000	.8440
.8000	1.2123	.8000	.8902
.9000	1.1708	.9000	.9322
1.0000	1.0848	1.0000	1.0848

TABLE III. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 13.500 DEG DYNAMIC PRESSURE = 21.147 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.1355	0.0000	2.1355
.0050	4.0449	.0050	.6930
.0100	4.2175	.0100	.2141
.0175	4.3156	.0175	.0535
.0500	4.1371	.0500	.1398
.0750	3.5631	.0750	.2915
.1000	3.3817	.1000	.3956
.2000	2.7868	.2000	.5786
.3000	2.3972	.3000	.6629
.4000	2.0403	.4000	.7331
.5000	1.5525	.5000	.7584
.6000	1.3265	.6000	.8104
.7000	1.2849	.7000	.8378
.8000	1.2670	.8000	.8911
.9000	1.2016	.9000	.9642
1.0000	1.1510	1.0000	1.1510

ALPHA = 14.520 DEG DYNAMIC PRESSURE = 21.010 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.4009	0.0000	2.4009
.0050	4.2809	.0050	.9071
.0100	4.2479	.0100	.4011
.0175	4.3527	.0175	.0689
.0500	4.2479	.0500	.0868
.0750	3.6792	.0750	.2485
.1000	3.4067	.1000	.3952
.2000	2.8080	.2000	.5548
.3000	2.3680	.3000	.6517
.4000	1.6136	.4000	.7358
.5000	1.6405	.5000	.7648
.6000	1.3411	.6000	.8079
.7000	1.2783	.7000	.8369
.8000	1.2843	.8000	.9061
.9000	1.2244	.9000	.9690
1.0000	1.1645	1.0000	1.1645

ALPHA = 15.550 DEG DYNAMIC PRESSURE = 20.973 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.6270	0.0000	2.6270
.0050	4.3904	.0050	1.0166
.0100	4.5193	.0100	.4168
.0175	4.4834	.0175	.1050
.0500	4.1805	.0500	.0900
.0750	3.6377	.0750	.1889
.1000	3.4098	.1000	.3179
.2000	2.7500	.2000	.5225
.3000	2.0393	.3000	.6337
.4000	1.4875	.4000	.7074
.5000	1.3915	.5000	.7505
.6000	1.3885	.6000	.8100
.7000	1.3255	.7000	.8475
.8000	1.3675	.8000	.9106
.9000	1.4035	.9000	.9764
1.0000	1.2415	1.0000	1.2415

TABLE III. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 16.580 DEG DYNAMIC PRESSURE = 21.138 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.7642	0.0000	2.7642
.0050	4.4543	.0050	1.1604
.0100	4.5317	.0100	.4820
.0175	4.5882	.0175	.1428
.0500	4.1627	.0500	.0744
.0750	3.6152	.0750	.1934
.1000	3.2909	.1000	.2916
.2000	2.4340	.2000	.5121
.3000	1.6276	.3000	.6295
.4000	1.3985	.4000	.7103
.5000	1.4699	.5000	.7426
.6000	1.4461	.6000	.8135
.7000	1.4074	.7000	.8543
.8000	1.4729	.8000	.9316
.9000	1.4758	.9000	1.0327
1.0000	1.2884	1.0000	1.2884

ALPHA = 17.580 DEG DYNAMIC PRESSURE = 21.019 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.9654	0.0000	2.9654
.0050	4.8416	.0050	1.2897
.0100	4.7788	.0100	.5536
.0175	4.7129	.0175	.1676
.0500	4.2850	.0500	.0718
.0750	3.4651	.0750	.2603
.1000	3.2198	.1000	.2992
.2000	2.2532	.2000	.4896
.3000	1.5470	.3000	.6118
.4000	1.4932	.4000	.7009
.5000	1.5141	.5000	.7510
.6000	1.5620	.6000	.8259
.7000	1.5680	.7000	.8881
.8000	1.6218	.8000	.9736
.9000	1.5051	.9000	1.0810
1.0000	1.3256	1.0000	1.3256

ALPHA = 18.580 DEG DYNAMIC PRESSURE = 21.046 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.9287	0.0000	2.9287
.0050	4.6710	.0050	1.2701
.0100	4.5694	.0100	.5469
.0175	4.3483	.0175	.1614
.0500	3.8223	.0500	.0448
.0750	2.7972	.0750	.1644
.1000	2.5940	.1000	.2660
.2000	1.5600	.2000	.4692
.3000	1.4883	.3000	.5941
.4000	1.4763	.4000	.6866
.5000	1.5451	.5000	.7409
.6000	1.5779	.6000	.8164
.7000	1.5660	.7000	.8855
.8000	1.5391	.8000	.9751
.9000	1.5510	.9000	1.1050
1.0000	1.3568	1.0000	1.3568

TABLE III. - CONCLUDED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 19.580 DEG DYNAMIC PRESSURE = 20.782 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.8903	0.0000	2.8903
.0050	4.3642	.0050	1.2590
.0100	4.2552	.0100	.5811
.0175	4.4610	.0175	.2119
.0500	3.5168	.0500	.0424
.0750	2.3304	.0750	.1574
.1000	1.9269	.1000	.2633
.2000	1.5495	.2000	.4680
.3000	1.5647	.3000	.5838
.4000	1.5949	.4000	.6796
.5000	1.6131	.5000	.7460
.6000	1.6767	.6000	.8089
.7000	1.6373	.7000	.8825
.8000	1.6192	.8000	.9825
.9000	1.5738	.9000	1.1047
1.0000	1.3861	1.0000	1.3861

ALPHA = 20.590 DEG DYNAMIC PRESSURE = 20.864 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.8669	0.0000	2.8669
.0050	4.3741	.0050	1.3234
.0100	4.1149	.0100	.5848
.0175	3.9220	.0175	.1839
.0500	2.4870	.0500	.0241
.0750	1.7816	.0750	.1296
.1000	1.6640	.1000	.2472
.2000	1.5736	.2000	.4498
.3000	1.5887	.3000	.5730
.4000	1.5947	.4000	.6648
.5000	1.6038	.5000	.7260
.6000	1.6158	.6000	.8071
.7000	1.6249	.7000	.8797
.8000	1.6550	.8000	.9822
.9000	1.5887	.9000	1.1203
1.0000	1.4289	1.0000	1.4289

TABLE IV

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -5.100 DEG DYNAMIC PRESSURE = 59.368 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.7289	0.0000	.7289
.0050	.0148	.0050	2.3466
.0100	.0021	.0100	3.0967
.0175	.0551	.0175	3.2207
.0500	.5901	.0500	2.7217
.0750	.8253	.0750	2.4791
.1000	1.0118	.1000	2.2852
.2000	1.3465	.2000	1.6044
.3000	1.4991	.3000	1.4049
.4000	1.4896	.4000	1.3091
.5000	1.4535	.5000	1.1845
.6000	1.3953	.6000	1.1160
.7000	1.3465	.7000	1.0449
.8000	1.1972	.8000	.9839
.9000	1.0022	.9000	.9169
1.0000	.8645	1.0000	.8645

ALPHA = -4.040 DEG DYNAMIC PRESSURE = 59.386 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.6143	0.0000	.6143
.0050	.0011	.0050	2.1712
.0100	.0328	.0100	2.5005
.0175	.1546	.0175	2.9083
.0500	.7848	.0500	2.5143
.0750	.9701	.0750	2.1987
.1000	1.1512	.1000	2.0504
.2000	1.4425	.2000	1.5467
.3000	1.5505	.3000	1.3667
.4000	1.5262	.4000	1.2399
.5000	1.4838	.5000	1.1366
.6000	1.4128	.6000	1.0764
.7000	1.3493	.7000	1.0204
.8000	1.2084	.8000	.9726
.9000	1.0072	.9000	.9101
1.0000	.8558	1.0000	.8558

ALPHA = -3.000 DEG DYNAMIC PRESSURE = 59.774 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.3262	0.0000	.3262
.0050	-.0074	.0050	1.7572
.0100	.0673	.0100	1.9087
.0175	.2020	.0175	2.4212
.0500	.8755	.0500	2.2307
.0750	1.1059	.0750	2.0245
.1000	1.2069	.1000	1.9477
.2000	1.5163	.2000	1.4477
.3000	1.6004	.3000	1.2920
.4000	1.5762	.4000	1.2040
.5000	1.5194	.5000	1.1101
.6000	1.4521	.6000	1.0597
.7000	1.3458	.7000	1.0016
.8000	1.2216	.8000	.9658
.9000	1.0143	.9000	.9087
1.0000	.8565	1.0000	.8565

Note: The conversion factor from pounds force/square foot to newtons/square meter is 47.880258.

TABLE IV. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -1.940 DEG DYNAMIC PRESSURE = 59.332 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.1972	0.0000	.1972
.0050	-.0011	.0050	1.3707
.0100	.1463	.0100	1.7406
.0175	.3106	.0175	1.8318
.0500	1.0251	.0500	1.9357
.0750	1.2477	.0750	1.9092
.1000	1.3601	.1000	1.7968
.2000	1.5943	.2000	1.3974
.3000	1.7173	.3000	1.2911
.4000	1.6378	.4000	1.1960
.5000	1.5731	.5000	1.1014
.6000	1.4746	.6000	1.0556
.7000	1.3738	.7000	.9960
.8000	1.2360	.8000	.9612
.9000	1.0272	.9000	.9072
1.0000	.8682	1.0000	.8682

ALPHA = -.900 DEG DYNAMIC PRESSURE = 59.892 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.1187	0.0000	.1187
.0050	.0452	.0050	.9220
.0100	.2919	.0100	1.4692
.0175	.4148	.0175	1.6824
.0500	1.1153	.0500	1.6519
.0750	1.3652	.0750	1.7937
.1000	1.5164	.1000	1.6676
.2000	1.6866	.2000	1.3286
.3000	1.7737	.3000	1.2014
.4000	1.6939	.4000	1.1335
.5000	1.5931	.5000	1.0608
.6000	1.4870	.6000	1.0224
.7000	1.3883	.7000	.9691
.8000	1.2434	.8000	.9411
.9000	1.0271	.9000	.8977
1.0000	.8632	1.0000	.8632

ALPHA = .160 DEG DYNAMIC PRESSURE = 59.341 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	.0223	0.0000	.0223
.0050	.1208	.0050	.6126
.0100	.3169	.0100	1.0355
.0175	.5268	.0175	1.3524
.0500	1.3821	.0500	1.5888
.0750	1.5220	.0750	1.5718
.1000	1.6768	.1000	1.5390
.2000	1.8252	.2000	1.2596
.3000	1.8633	.3000	1.1678
.4000	1.7170	.4000	1.1317
.5000	1.6238	.5000	1.0577
.6000	1.5125	.6000	1.0079
.7000	1.4097	.7000	.9649
.8000	1.2571	.8000	.9383
.9000	1.0281	.9000	.8998
1.0000	.8617	1.0000	.8617

TABLE IV. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 1.250 DEG DYNAMIC PRESSURE = 59.648 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.0095	0.0000	.0095
.0050	.2235	.0050	.4439
.0100	.5209	.0100	.6843
.0175	.8562	.0175	1.0766
.0500	1.6133	.0500	1.4499
.0750	1.6481	.0750	1.3961
.1000	1.7219	.1000	1.4277
.2000	1.9223	.2000	1.1675
.3000	1.9223	.3000	1.1145
.4000	1.7736	.4000	1.0759
.5000	1.6565	.5000	.9957
.6000	1.5427	.6000	.9805
.7000	1.4383	.7000	.9435
.8000	1.2622	.8000	.9290
.9000	1.0302	.9000	.8867
1.0000	.8689	1.0000	.8689

ALPHA = 2.340 DEG DYNAMIC PRESSURE = 59.287 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	-.0032	0.0000	-.0032
.0050	.3098	.0050	.3098
.0100	.6662	.0100	.5633
.0175	1.0545	.0175	.8317
.0500	1.8024	.0500	1.2593
.0750	1.8289	.0750	1.3059
.1000	1.9053	.1000	1.2550
.2000	1.9838	.2000	1.1603
.3000	1.9775	.3000	1.0990
.4000	1.8449	.4000	1.0629
.5000	1.6486	.5000	.9815
.6000	1.5457	.6000	.9677
.7000	1.4226	.7000	.9339
.8000	1.2614	.8000	.9237
.9000	1.0237	.9000	.8876
1.0000	.8604	1.0000	.8604

ALPHA = 3.430 DEG DYNAMIC PRESSURE = 59.133 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.0160	0.0000	.0160
.0050	.5414	.0050	.1213
.0100	.9668	.0100	.3829
.0175	1.3253	.0175	.6180
.0500	2.1581	.0500	1.0870
.0750	2.0177	.0750	1.1551
.1000	2.0869	.1000	1.1838
.2000	2.1060	.2000	1.0607
.3000	2.0411	.3000	1.0355
.4000	1.8975	.4000	1.0064
.5000	1.7082	.5000	.9492
.6000	1.5848	.6000	.9291
.7000	1.4529	.7000	.9148
.8000	1.2721	.8000	.9004
.9000	1.0381	.9000	.8799
1.0000	.8679	1.0000	.8679

TABLE IV. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 4.510 DEG DYNAMIC PRESSURE = 59.332 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.1431	0.0000	.1431
.0050	.7208	.0050	.0477
.0100	1.3018	.0100	.1887
.0175	1.3770	.0175	.4198
.0500	2.1816	.0500	.9678
.0750	2.1180	.0750	.9731
.1000	2.2113	.1000	1.0463
.2000	2.1975	.2000	.9637
.3000	2.1191	.3000	1.0025
.4000	1.9071	.4000	.9685
.5000	1.7470	.5000	.9437
.6000	1.6092	.6000	.9285
.7000	1.4491	.7000	.9149
.8000	1.2668	.8000	.9047
.9000	1.0060	.9000	.8832
1.0000	.8746	1.0000	.8746

ALPHA = 5.590 DEG DYNAMIC PRESSURE = 59.540 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.1870	0.0000	.1870
.0050	1.0321	.0050	.0074
.0100	1.4483	.0100	.1151
.0175	1.8772	.0175	.2609
.0500	2.3705	.0500	.7754
.0750	2.3166	.0750	.9021
.1000	2.4191	.1000	.9824
.2000	2.2986	.2000	.9479
.3000	2.1592	.3000	.9479
.4000	1.9162	.4000	.9457
.5000	1.7641	.5000	.9013
.6000	1.6152	.6000	.9100
.7000	1.4514	.7000	.8935
.8000	1.2507	.8000	.8911
.9000	1.0109	.9000	.8848
1.0000	.8831	1.0000	.8831

ALPHA = 6.710 DEG DYNAMIC PRESSURE = 59.206 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.3176	0.0000	.3176
.0050	1.4033	.0050	-.0011
.0100	1.6923	.0100	.0297
.0175	2.1576	.0175	.1434
.0500	2.5251	.0500	.6225
.0750	2.4667	.0750	.8382
.1000	2.6006	.1000	.8711
.2000	2.3817	.2000	.8913
.3000	2.2245	.3000	.9074
.4000	1.9727	.4000	.9217
.5000	1.7879	.5000	.8825
.6000	1.6232	.6000	.9056
.7000	1.4533	.7000	.8828
.8000	1.2429	.8000	.8916
.9000	1.0230	.9000	.8883
1.0000	.9157	1.0000	.9157

TABLE IV. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 7.760 DEG DYNAMIC PRESSURE = 59.016 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.5009	0.0000	.5009
.0050	1.6892	.0050	.0021
.0100	2.1720	.0100	.0075
.0175	2.3457	.0175	.0810
.0500	2.8775	.0500	.5584
.0750	2.6697	.0750	.7577
.1000	2.7049	.1000	.8291
.2000	2.4235	.2000	.8376
.3000	2.2562	.3000	.8703
.4000	2.0228	.4000	.8851
.5000	1.8214	.5000	.8655
.6000	1.7265	.6000	.8796
.7000	1.4547	.7000	.8733
.8000	1.2309	.8000	.8879
.9000	1.0189	.9000	.8871
1.0000	.9400	1.0000	.9400

ALPHA = 8.810 DEG DYNAMIC PRESSURE = 59.359 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.6930	0.0000	.6930
.0050	1.9867	.0050	.0265
.0100	2.4699	.0100	-.0021
.0175	2.6776	.0175	.0254
.0500	3.0866	.0500	.4228
.0750	2.8884	.0750	.5891
.1000	2.8365	.1000	.6877
.2000	2.5303	.2000	.7752
.3000	2.3057	.3000	.8300
.4000	2.0291	.4000	.8470
.5000	1.8225	.5000	.8377
.6000	1.6159	.6000	.8577
.7000	1.4400	.7000	.8605
.8000	1.2153	.8000	.8853
.9000	1.0416	.9000	.9015
1.0000	.9769	1.0000	.9769

ALPHA = 9.900 DEG DYNAMIC PRESSURE = 59.458 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	.9774	0.0000	.9774
.0050	2.6012	.0050	.1618
.0100	2.6805	.0100	.0138
.0175	2.9228	.0175	.0021
.0500	3.3364	.0500	.3406
.0750	3.0539	.0750	.5353
.1000	2.8995	.1000	.6072
.2000	2.5980	.2000	.7298
.3000	2.3642	.3000	.7759
.4000	2.0522	.4000	.8323
.5000	1.8237	.5000	.8234
.6000	1.5931	.6000	.8438
.7000	1.3995	.7000	.8508
.8000	1.1795	.8000	.8808
.9000	1.1075	.9000	.9118
1.0000	1.0546	1.0000	1.0546

TABLE IV. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 10.950 DEG DYNAMIC PRESSURE = 59.115 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	1.1395	0.0000	1.1395
.0050	2.8599	.0050	.2671
.0100	3.0961	.0100	.0521
.0175	3.3962	.0175	.0021
.0500	3.7419	.0500	.3245
.0750	3.2163	.0750	.4745
.1000	3.1451	.1000	.5543
.2000	2.6556	.2000	.7077
.3000	2.3833	.3000	.7553
.4000	2.0481	.4000	.8163
.5000	1.8034	.5000	.8078
.6000	1.5491	.6000	.8324
.7000	1.3204	.7000	.8520
.8000	1.1618	.8000	.8914
.9000	1.1331	.9000	.9256
1.0000	1.0703	1.0000	1.0703

ALPHA = 12.000 DEG DYNAMIC PRESSURE = 59.142 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	1.5059	0.0000	1.5059
.0050	3.0500	.0050	.3914
.0100	3.5807	.0100	.1244
.0175	3.8455	.0175	.0043
.0500	3.8040	.0500	.2191
.0750	3.3489	.0750	.3892
.1000	3.2500	.1000	.5232
.2000	2.7427	.2000	.6571
.3000	2.4024	.3000	.7447
.4000	2.0451	.4000	.7813
.5000	1.7739	.5000	.7896
.6000	1.4984	.6000	.8273
.7000	1.3198	.7000	.8501
.8000	1.2017	.8000	.9003
.9000	1.1624	.9000	.9518
1.0000	1.1369	1.0000	1.1369

ALPHA = 13.070 DEG DYNAMIC PRESSURE = 59.106 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C'_P	X/C	C'_P
0.0000	1.9207	0.0000	1.9207
.0050	3.5723	.0050	.5438
.0100	3.8809	.0100	.1915
.0175	3.9096	.0175	.0074
.0500	3.9415	.0500	.1969
.0750	3.4658	.0750	.3639
.1000	3.2881	.1000	.4437
.2000	2.7305	.2000	.6040
.3000	2.3932	.3000	.6985
.4000	1.8963	.4000	.7547
.5000	1.6004	.5000	.7828
.6000	1.3227	.6000	.8232
.7000	1.2684	.7000	.8529
.8000	1.2312	.8000	.9227
.9000	1.2503	.9000	.9735
1.0000	1.1833	1.0000	1.1833

TABLE IV. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 14.140 DEG DYNAMIC PRESSURE = 59.178 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.0757	0.0000	2.0757
.0050	3.8294	.0050	.7068
.0100	3.9792	.0100	.2296
.0175	4.0749	.0175	.0223
.0500	4.0196	.0500	.1435
.0750	3.4914	.0750	.2944
.1000	3.2661	.1000	.4113
.2000	2.6390	.2000	.5849
.3000	2.2298	.3000	.6861
.4000	1.5581	.4000	.7590
.5000	1.3944	.5000	.7869
.6000	1.3849	.6000	.8383
.7000	1.4380	.7000	.8629
.8000	1.3827	.8000	.9331
.9000	1.2786	.9000	1.0104
1.0000	1.3094	1.0000	1.3094

ALPHA = 15.120 DEG DYNAMIC PRESSURE = 58.727 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.1141	0.0000	2.1141
.0050	3.9391	.0050	.7968
.0100	4.1372	.0100	.2635
.0175	4.1683	.0175	.0278
.0500	3.9862	.0500	.1146
.0750	3.4047	.0750	.2806
.1000	3.2783	.1000	.3802
.2000	2.4868	.2000	.5745
.3000	1.7982	.3000	.6693
.4000	1.4276	.4000	.7429
.5000	1.4341	.5000	.7706
.6000	1.4587	.6000	.8359
.7000	1.4694	.7000	.8844
.8000	1.4630	.8000	.9395
.9000	1.4308	.9000	1.0417
1.0000	1.3419	1.0000	1.3419

ALPHA = 16.120 DEG DYNAMIC PRESSURE = 57.815 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.4031	0.0000	2.4031
.0050	4.1002	.0050	.8311
.0100	4.1981	.0100	.3514
.0175	4.2330	.0175	.0674
.0500	3.9153	.0500	.1033
.0750	3.2985	.0750	.2437
.1000	3.1375	.1000	.3623
.2000	2.1355	.2000	.5402
.3000	1.4784	.3000	.6606
.4000	1.4273	.4000	.7369
.5000	1.4447	.5000	.7746
.6000	1.4719	.6000	.8473
.7000	1.5274	.7000	.8943
.8000	1.5339	.8000	.9603
.9000	1.5056	.9000	1.0607
1.0000	1.3653	1.0000	1.3653

TABLE IV. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 17.150 DEG DYNAMIC PRESSURE = 59.684 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.3837	0.0000	2.3837
.0050	4.0530	.0050	.8879
.0100	4.0825	.0100	.3414
.0175	4.1721	.0175	.0696
.0500	4.0951	.0500	.0980
.0750	2.7536	.0750	.2224
.1000	2.7926	.1000	.3362
.2000	2.0370	.2000	.5223
.3000	1.4753	.3000	.6345
.4000	1.4764	.4000	.7206
.5000	1.5344	.5000	.7682
.6000	1.5733	.6000	.8307
.7000	1.5639	.7000	.8874
.8000	1.5354	.8000	.9690
.9000	1.5502	.9000	1.0663
1.0000	1.4227	1.0000	1.4227

ALPHA = 18.150 DEG DYNAMIC PRESSURE = 59.233 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.3169	0.0000	2.3169
.0050	3.8916	.0050	.9334
.0100	3.8831	.0100	.3600
.0175	3.7547	.0175	.0807
.0500	3.3554	.0500	.0913
.0750	2.6886	.0750	.2134
.1000	2.0504	.1000	.3355
.2000	1.5460	.2000	.5117
.3000	1.5450	.3000	.6235
.4000	1.5577	.4000	.7153
.5000	1.5853	.5000	.7575
.6000	1.6182	.6000	.8325
.7000	1.6257	.7000	.8879
.8000	1.6639	.8000	.9869
.9000	1.5928	.9000	1.1042
1.0000	1.4430	1.0000	1.4430

ALPHA = 19.170 DEG DYNAMIC PRESSURE = 60.424 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.2890	0.0000	2.2890
.0050	3.8014	.0050	.9410
.0100	3.7296	.0100	.3872
.0175	3.5911	.0175	.0906
.0500	3.0790	.0500	.0739
.0750	1.7956	.0750	.2019
.1000	1.5926	.1000	.3050
.2000	1.5187	.2000	.4908
.3000	1.5093	.3000	.6127
.4000	1.5895	.4000	.7009
.5000	1.5811	.5000	.7443
.6000	1.5791	.6000	.8247
.7000	1.6020	.7000	.8807
.8000	1.6301	.8000	.9761
.9000	1.5562	.9000	1.0898
1.0000	1.4354	1.0000	1.4354

TABLE IV. - CONCLUDED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 20.190 DEG DYNAMIC PRESSURE = 59.820 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.2585	0.0000	2.2585
.0050	3.5727	.0050	.9631
.0100	3.5633	.0100	.4153
.0175	3.3414	.0175	.1093
.0500	2.3930	.0500	.0599
.0750	1.6707	.0750	.1745
.1000	1.5940	.1000	.2702
.2000	1.5940	.2000	.4704
.3000	1.5855	.3000	.5864
.4000	1.6013	.4000	.6859
.5000	1.6181	.5000	.7326
.6000	1.6644	.6000	.8127
.7000	1.6444	.7000	.8794
.8000	1.6654	.8000	.9877
.9000	1.6360	.9000	1.1063
1.0000	1.5140	1.0000	1.5140

ALPHA = 21.220 DEG DYNAMIC PRESSURE = 59.278 LBF/SQ.FT. FLAP = 0 DEG

WING UPPER SURFACE		WING LOWER SURFACE	
X/C	C_P'	X/C	C_P'
0.0000	2.4531	0.0000	2.4531
.0050	3.7083	.0050	1.0112
.0100	3.5173	.0100	.4531
.0175	3.3529	.0175	.1369
.0500	2.0064	.0500	.0499
.0750	1.6478	.0750	.1634
.1000	1.6319	.1000	.2578
.2000	1.6096	.2000	.4589
.3000	1.6223	.3000	.5792
.4000	1.6425	.4000	.6894
.5000	1.6817	.5000	.7552
.6000	1.6966	.6000	.8264
.7000	1.7061	.7000	.9030
.8000	1.6690	.8000	1.0070
.9000	1.6457	.9000	1.1362
1.0000	1.5268	1.0000	1.5268

TABLE V

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = .070 DEG DYNAMIC PRESSURE = 4.535 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	.5625	0.0000	.5625				
.0050	1.5078	.0050	.0200				
.0100	1.9205	.0100	-.0033	.0100	2.9003	.0100	.4896
.0175	2.1036	.0175	.0599				
.0500	2.5762	.0500	.4560	.0500	2.8666	.0500	.0338
.0750	2.4864	.0750	.5692				
.1000	2.5430	.1000	.6191	.1000	2.4209	.1000	.0743
.2000	2.4664	.2000	.5943				
.3000	2.3899	.3000	.5943	.3000	1.9212	.3000	.2566
.4000	2.2301	.4000	.5672				
.5000	2.1635	.5000	.5436	.5000	1.2425	.5000	.3748
.6000	2.0936	.6000	.5132				
.7000	2.0071	.7000	.5031	.7000	1.0771	.7000	.5065
.8000	1.9039	.8000	.3545				
.9000	1.7541	.9000	.3005	.9000	1.0906	.8792	.6550
1.0000	2.1502	1.0000	2.1502	1.0000	1.0366	1.0000	1.0366

ALPHA = .190 DEG DYNAMIC PRESSURE = 9.861 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	.4761	0.0000	.4761				
.0050	1.4925	.0050	.0475				
.0100	1.7068	.0100	.0153	.0100	2.9995	.0100	.4596
.0175	1.9364	.0175	.0704				
.0500	2.5747	.0500	.4562	.0500	2.9622	.0500	.0264
.0750	2.6099	.0750	.5694				
.1000	2.6604	.1000	.6261	.1000	2.8704	.1000	.0683
.2000	2.4997	.2000	.6071				
.3000	2.4354	.3000	.5947	.3000	1.9742	.3000	.2609
.4000	2.3022	.4000	.5807				
.5000	2.1936	.5000	.5419	.5000	1.2267	.5000	.3665
.6000	2.1063	.6000	.5124				
.7000	1.9946	.7000	.4410	.7000	1.0730	.7000	.4922
.8000	1.9119	.8000	.3556				
.9000	1.8124	.9000	.2950	.9000	1.0792	.8792	.6413
1.0000	2.2364	1.0000	2.2364	1.0000	1.0342	1.0000	1.0342

ALPHA = .380 DEG DYNAMIC PRESSURE = 15.620 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	.4832	0.0000	.4832				
.0050	1.5539	.0050	.1034				
.0100	2.1511	.0100	-.0029	.0100	3.3698	.0100	.4598
.0175	2.4266	.0175	.0338				
.0500	2.9194	.0500	.4136	.0500	3.3256	.0500	.0069
.0750	2.7909	.0750	.5731				
.1000	2.8875	.1000	.6069	.1000	3.2078	.1000	.0598
.2000	2.7638	.2000	.6676				
.3000	2.6739	.3000	.6529	.3000	2.1755	.3000	.2617
.4000	2.4710	.4000	.6039				
.5000	2.4121	.5000	.5695	.5000	1.3564	.5000	.3892
.6000	2.2613	.6000	.5499				
.7000	2.2130	.7000	.4509	.7000	1.2002	.7000	.5097
.8000	2.1231	.8000	.3804				
.9000	1.9733	.9000	.3186	.9000	1.1894	.8792	.6715
1.0000	2.4555	1.0000	2.4555	1.0000	1.1187	1.0000	1.1187

Note: The conversion factor from pounds force/square foot to newtons/square meter is 47.880258.

TABLE V. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = .540 DEG DYNAMIC PRESSURE = 21.384 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	.6777	0.0000	.6777				
.0050	1.7167	.0050	.0741				
.0100	2.2659	.0100	.0064	.0100	3.3812	.0100	.4504
.0175	2.4600	.0175	.0254				
.0500	3.1073	.0500	.4525	.0500	3.3109	.0500	.0057
.0750	2.9873	.0750	.5704				
.1000	2.8581	.1000	.6402	.1000	3.2062	.1000	.0666
.2000	2.7219	.2000	.6230				
.3000	2.6626	.3000	.6280	.3000	2.1566	.3000	.2678
.4000	2.5115	.4000	.5972				
.5000	2.4014	.5000	.5621	.5000	1.3337	.5000	.3874
.6000	2.2504	.6000	.5256				
.7000	2.2221	.7000	.4375	.7000	1.2074	.7000	.5170
.8000	2.1247	.8000	.3752				
.9000	1.9871	.9000	.3201	.9000	1.1802	.8792	.6731
1.0000	2.4494	1.0000	2.4494	1.0000	1.1084	1.0000	1.1084

ALPHA = .790 DEG DYNAMIC PRESSURE = 31.273 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	.7095	0.0000	.7095				
.0050	1.9220	.0050	.0574				
.0100	2.2444	.0100	-.0024	.0100	3.3697	.0100	.4279
.0175	2.5273	.0175	.0333				
.0500	3.3005	.0500	.4185	.0500	3.3383	.0500	-.0039
.0750	2.9795	.0750	.5560				
.1000	2.9081	.1000	.6444	.1000	3.2049	.1000	.0597
.2000	2.7932	.2000	.6127				
.3000	2.6822	.3000	.6024	.3000	2.2718	.3000	.2624
.4000	2.5157	.4000	.5897				
.5000	2.3873	.5000	.5465	.5000	1.5953	.5000	.3848
.6000	2.2990	.6000	.5004				
.7000	2.2290	.7000	.4338	.7000	1.2230	.7000	.5073
.8000	2.1344	.8000	.3785				
.9000	1.9973	.9000	.3016	.9000	1.1646	.8792	.6505
1.0000	2.4703	1.0000	2.4703	1.0000	1.0513	1.0000	1.0513

TABLE V. - CONCLUDED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 1.030 DEG DYNAMIC PRESSURE = 41.462 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	.6888	0.0000	.6888				
.0050	2.1556	.0050	.0637				
.0100	2.2073	.0100	.0004	.0100	3.3183	.0100	.4014
.0175	2.6034	.0175	.0171				
.0500	3.0956	.0500	.3906	.0500	3.2735	.0500	-.0022
.0750	3.0687	.0750	.5505				
.1000	2.9988	.1000	.5923	.1000	3.1599	.1000	.0665
.2000	2.8255	.2000	.6131				
.3000	2.6908	.3000	.5857	.3000	2.2212	.3000	.2666
.4000	2.4978	.4000	.5928				
.5000	2.3726	.5000	.5413	.5000	1.5622	.5000	.3926
.6000	2.2779	.6000	.5117				
.7000	2.2164	.7000	.4303	.7000	1.2506	.7000	.5121
.8000	2.1072	.8000	.3748				
.9000	1.9856	.9000	.3006	.9000	1.2018	.8792	.6531
1.0000	2.4268	1.0000	2.4268	1.0000	1.0679	1.0000	1.0679

ALPHA = 1.240 DEG DYNAMIC PRESSURE = 48.461 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	.6096	0.0000	.6096				
.0050	1.8340	.0050	.0486				
.0100	2.4672	.0100	-.0016	.0100	3.3237	.0100	.3967
.0175	2.7027	.0175	.0140				
.0500		.0500	.3741	.0500	3.2898	.0500	-.0009
.0750		.0750	.5479				
.1000		.1000	.5641	.1000	3.1534	.1000	.0686
.2000	2.8367	.2000	.6385				
.3000	2.7460	.3000	.6246	.3000	2.2075	.3000	.2679
.4000	2.5059	.4000	.5942				
.5000	2.4000	.5000	.5543	.5000	1.5259	.5000	.3922
.6000	2.2965	.6000	.5160				
.7000	2.2237	.7000	.4400	.7000	1.2679	.7000	.5103
.8000	2.1174	.8000	.3773				
.9000	1.9876	.9000	.3055	.9000	1.2378	.8792	.6585
1.0000	2.4143	1.0000	2.4143	1.0000	1.1184	1.0000	1.1184

TABLE VI

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -4.800 DEG DYNAMIC PRESSURE = 8.840 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C'_P	X/C	C'_P	X/C	C'_P	X/C	C'_P
0.0000	-.0034	0.0000	-.0034				
.0050	.2682	.0050	.2785				
.0100	.5911	.0100	.6321	.0100	3.1944	.0100	.6169
.0175	.9003	.0175	.7927				
.0500	1.6944	.0500	1.1258	.0500	3.1615	.0500	.0295
.0750	1.7255	.0750	1.1173				
.1000	1.9390	.1000	1.1122	.1000	3.0937	.1000	.0641
.2000	2.1287	.2000	.9357				
.3000	2.2260	.3000	.8889	.3000	2.1788	.3000	.3067
.4000	2.1782	.4000	.7711				
.5000	2.1252	.5000	.6931	.5000	1.3689	.5000	.4419
.6000	2.0825	.6000	.6221				
.7000	2.0586	.7000	.5441	.7000	1.2199	.7000	.5805
.8000	2.0227	.8000	.4575				
.9000	1.9390	.9000	.3829	.9000	1.2285	.8792	.7503
1.0000	2.3866	1.0000	2.3866	1.0000	1.1731	1.0000	1.1731

ALPHA = -3.860 DEG DYNAMIC PRESSURE = 8.931 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C'_P	X/C	C'_P	X/C	C'_P	X/C	C'_P
0.0000	.0220	0.0000	.0220				
.0050	.4667	.0050	.1032				
.0100	.7779	.0100	.3467	.0100	3.2409	.0100	.5626
.0175	1.1380	.0175	.5462				
.0500	2.0326	.0500	.9588	.0500	3.1911	.0500	.0172
.0750	2.1222	.0750	1.0044				
.1000	2.1662	.1000	1.0197	.1000	3.1310	.1000	.0652
.2000	2.2507	.2000	.8541				
.3000	2.3403	.3000	.8233	.3000	2.1772	.3000	.2899
.4000	2.2473	.4000	.7547				
.5000	2.1898	.5000	.6586	.5000	1.3738	.5000	.4305
.6000	2.1205	.6000	.5831				
.7000	2.1036	.7000	.5111	.7000	1.2006	.7000	.5643
.8000	2.0630	.8000	.4356				
.9000	1.9785	.9000	.3756	.9000	1.1954	.8792	.7392
1.0000	2.4063	1.0000	2.4063	1.0000	1.1371	1.0000	1.1371

ALPHA = -2.850 DEG DYNAMIC PRESSURE = 8.968 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C'_P	X/C	C'_P	X/C	C'_P	X/C	C'_P
0.0000	.0286	0.0000	.0286				
.0050	.5726	.0050	.0269				
.0100	.9313	.0100	.1667	.0100	3.2806	.0100	.5346
.0175	1.2344	.0175	.3536				
.0500	2.1353	.0500	.7679	.0500	3.2344	.0500	.0137
.0750	2.2010	.0750	.8521				
.1000	2.1774	.1000	.9009	.1000	3.1608	.1000	.0683
.2000	2.3745	.2000	.8677				
.3000	2.3745	.3000	.8182	.3000	2.2555	.3000	.2801
.4000	2.3223	.4000	.7686				
.5000	2.2482	.5000	.6713	.5000	1.4126	.5000	.4133
.6000	2.1909	.6000	.5910				
.7000	2.1572	.7000	.5244	.7000	1.2076	.7000	.5551
.8000	2.0932	.8000	.4663				
.9000	1.9703	.9000	.3706	.9000	1.2093	.8792	.7174
1.0000	2.4452	1.0000	2.4452	1.0000	1.1410	1.0000	1.1410

Note: The conversion factor from pounds force/square foot to newtons/square meter is 47.880258.

TABLE VI. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -1.830 DEG DYNAMIC PRESSURE = 8.913 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P	X/C	C_P	X/C	C_P	X/C	C_P
0.0000	.2609	0.0000	.2609				
.0050	.9794	.0050	0.0000				
.0100	1.1742	.0100	.0898	.0100	3.3284	.0100	.5207
.0175	1.5707	.0175	.2457				
.0500	2.4281	.0500	.7354	.0500	3.2699	.0500	.0137
.0750	2.3993	.0750	.7845				
.1000	2.5331	.1000	.7777	.1000	3.1889	.1000	.0619
.2000	2.5196	.2000	.7167				
.3000	2.4942	.3000	.7270	.3000	2.2109	.3000	.2818
.4000	2.3772	.4000	.7132				
.5000	2.3044	.5000	.6651	.5000	1.4385	.5000	.4142
.6000	2.2095	.6000	.5654				
.7000	2.1807	.7000	.4795	.7000	1.1979	.7000	.5396
.8000	2.1078	.8000	.4125				
.9000	1.9333	.9000	.3454	.9000	1.1875	.8792	.6977
1.0000	2.4298	1.0000	2.4298	1.0000	1.1343	1.0000	1.1343

ALPHA = -.850 DEG DYNAMIC PRESSURE = 8.904 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P	X/C	C_P	X/C	C_P	X/C	C_P
0.0000	.3986	0.0000	.3986				
.0050	1.3043	.0050	.0119				
.0100	1.6164	.0100	.0339	.0100	3.3180	.0100	.5178
.0175	1.8946	.0175	.1476				
.0500	2.6493	.0500	.6089	.0500	3.2628	.0500	.0155
.0750	2.6341	.0750	.7429				
.1000	2.6799	.1000	.7259	.1000	3.1611	.1000	.0602
.2000	2.6510	.2000	.7036				
.3000	2.6205	.3000	.6881	.3000	2.1925	.3000	.2873
.4000	2.4645	.4000	.6692				
.5000	2.3746	.5000	.6348	.5000	1.3573	.5000	.4215
.6000	2.2745	.6000	.5884				
.7000	2.2287	.7000	.4679	.7000	1.2077	.7000	.5402
.8000	2.1252	.8000	.4008				
.9000	1.9607	.9000	.3251	.9000	1.1991	.8792	.7053
1.0000	2.4746	1.0000	2.4746	1.0000	1.1423	1.0000	1.1423

ALPHA = .160 DEG DYNAMIC PRESSURE = 8.895 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P	X/C	C_P	X/C	C_P	X/C	C_P
0.0000	.4567	0.0000	.4567				
.0050	1.6112	.0050	.0441				
.0100	1.9712	.0100	.0017	.0100	3.3230	.0100	.5149
.0175	2.1936	.0175	.0374				
.0500	2.8812	.0500	.4958	.0500	3.2644	.0500	.0258
.0750	2.7013	.0750	.5314				
.1000	2.7878	.1000	.5739	.1000	3.1609	.1000	.0827
.2000	2.7437	.2000	.7026				
.3000	2.6792	.3000	.6733	.3000	2.1964	.3000	.2859
.4000	2.5026	.4000	.6389				
.5000	2.3990	.5000	.6079	.5000	1.3966	.5000	.4081
.6000	2.3023	.6000	.5786				
.7000	2.2038	.7000	.4701	.7000	1.2020	.7000	.5304
.8000	2.1002	.8000	.4150				
.9000	1.9831	.9000	.3358	.9000	1.1968	.8792	.6992
1.0000	2.4483	1.0000	2.4483	1.0000	1.1469	1.0000	1.1469

TABLE VI. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 1.150 DEG DYNAMIC PRESSURE = 8.895 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	.8184	0.0000	.8184				
.0050	2.0085	.0050	.0458				
.0100	2.4585	.0100	-.0034	.0100	3.2748	.0100	.4856
.0175	2.5943	.0175	.0136				
.0500	3.2021	.0500	.4109	.0500	3.2040	.0500	.0138
.0750	3.0782	.0750	.5178				
.1000	3.0595	.1000	.5535	.1000	3.1178	.1000	.0499
.2000	2.7895	.2000	.6131				
.3000	2.7386	.3000	.6165	.3000	2.1636	.3000	.2652
.4000	2.6130	.4000	.5907				
.5000	2.4415	.5000	.5528	.5000	1.3346	.5000	.0978
.6000	2.2564	.6000	.5218				
.7000	2.1783	.7000	.4787	.7000	1.1538	.7000	.5201
.8000	2.1053	.8000	.3685				
.9000	1.9729	.9000	.3100	.9000	1.1641	.8792	.6785
1.0000	2.4228	1.0000	2.4228	1.0000	1.1004	1.0000	1.1004

ALPHA = 2.170 DEG DYNAMIC PRESSURE = 8.840 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	1.1908	0.0000	1.1908				
.0050	2.4259	.0050	.3143				
.0100	2.9521	.0100	.0598	.0100	3.3264	.0100	.4800
.0175	2.9504	.0175	-.0120				
.0500	3.4629	.0500	.2255	.0500	3.2517	.0500	.0052
.0750	3.0905	.0750	.4100				
.1000	3.1161	.1000	.5296	.1000	3.1128	.1000	.0503
.2000	2.9282	.2000	.5614				
.3000	2.8189	.3000	.5874	.3000	2.1163	.3000	.2547
.4000	2.6839	.4000	.5666				
.5000	2.4276	.5000	.5372	.5000	1.3516	.5000	.3847
.6000	2.3285	.6000	.5008				
.7000	2.2585	.7000	.4938	.7000	1.1679	.7000	.5250
.8000	2.1457	.8000	.3587				
.9000	2.0091	.9000	.3015	.9000	1.1800	.8792	.6706
1.0000	2.4123	1.0000	2.4123	1.0000	1.0917	1.0000	1.0917

ALPHA = 3.180 DEG DYNAMIC PRESSURE = 8.830 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	1.3460	0.0000	1.3460				
.0050	2.5946	.0050	.3404				
.0100	3.0974	.0100	.0479	.0100	3.2989	.0100	.4996
.0175	3.2308	.0175	.0188				
.0500	3.5660	.0500	.2600	.0500	3.2120	.0500	.0035
.0750	3.2137	.0750	.3797				
.1000	3.2445	.1000	.4310	.1000	3.0677	.1000	.0659
.2000	3.1265	.2000	.5447				
.3000	2.9195	.3000	.5621	.3000	2.1100	.3000	.2741
.4000	2.7485	.4000	.5516				
.5000	2.4526	.5000	.5274	.5000	1.3323	.5000	.4007
.6000	2.3312	.6000	.4944				
.7000	2.2491	.7000	.4857	.7000	1.1519	.7000	.5343
.8000	2.1465	.8000	.3643				
.9000	2.0131	.9000	.2949	.9000	1.1762	.8792	.6783
1.0000	2.4509	1.0000	2.4509	1.0000	1.1241	1.0000	1.1241

TABLE VI. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 4.210 DEG DYNAMIC PRESSURE = 8.885 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	2.1961	0.0000	2.1961				
.0050	3.5831	.0050	.5949				
.0100	3.5508	.0100	.0952	.0100	3.2543	.0100	.5000
.0175	3.8023	.0175	.0221				
.0500	3.9927	.0500	.1870	.0500	3.1765	.0500	.0310
.0750	3.8839	.0750	.2907				
.1000	3.5984	.1000	.3501	.1000	3.0833	.1000	.0759
.2000	3.2873	.2000	.4586				
.3000	3.0867	.3000	.4982	.3000	2.1332	.3000	.2689
.4000	2.7332	.4000	.5310				
.5000	2.5207	.5000	.4896	.5000	1.3654	.5000	.3776
.6000	2.3966	.6000	.4724				
.7000	2.3185	.7000	.4396	.7000	1.1930	.7000	.5000
.8000	2.1944	.8000	.4069				
.9000	2.0431	.9000	.3017	.9000	1.1861	.8792	.6517
1.0000	2.4051	1.0000	2.4051	1.0000	1.1275	1.0000	1.1275

ALPHA = 5.220 DEG DYNAMIC PRESSURE = 8.803 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	2.7226	0.0000	2.7226				
.0050	4.1397	.0050	.6416				
.0100	4.2272	.0100	.2247	.0100	3.2323	.0100	.4698
.0175	4.1260	.0175	.0274				
.0500	4.4588	.0500	.1081	.0500	3.1224	.0500	.0070
.0750	4.0144	.0750	.2573				
.1000	3.8686	.1000	.3380	.1000	2.9934	.1000	.0557
.2000	3.3745	.2000	.4124				
.3000	3.1121	.3000	.4646	.3000	2.0956	.3000	.0199
.4000	2.7861	.4000	.4768				
.5000	2.5871	.5000	.4646	.5000	1.3416	.5000	.3706
.6000	2.4293	.6000	.4176				
.7000	2.3160	.7000	.4054	.7000	1.1589	.7000	.4942
.8000	2.1977	.8000	.3967				
.9000	2.0415	.9000	.2732	.9000	1.1537	.8792	.6456
1.0000	2.3898	1.0000	2.3898	1.0000	1.0910	1.0000	1.0910

ALPHA = 6.260 DEG DYNAMIC PRESSURE = 8.977 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	2.8785	0.0000	2.8785				
.0050	4.6348	.0050	1.0447				
.0100	4.6045	.0100	.4021	.0100	3.1867	.0100	.4385
.0175	4.9040	.0175	.1581				
.0500	4.7290	.0500	.0555	.0500	3.1491	.0500	.0051
.0750	4.2159	.0750	.1598				
.1000	4.0931	.1000	.2456	.1000	2.9371	.1000	.0546
.2000	3.3916	.2000	.4163				
.3000	3.1409	.3000	.4488	.3000	2.0550	.3000	.2406
.4000	2.8246	.4000	.4675				
.5000	2.6261	.5000	.4471	.5000	1.4146	.5000	.3617
.6000	2.4545	.6000	.4163				
.7000	2.3233	.7000	.3993	.7000	1.1893	.7000	.4795
.8000	2.1534	.8000	.3856				
.9000	2.0070	.9000	.2713	.9000	1.0426	.8792	.6382
1.0000	2.2964	1.0000	2.2964	1.0000	.9863	1.0000	.9863

TABLE VI. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 7.270 DEG DYNAMIC PRESSURE = 8.895 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	2.7488	0.0000	2.7488				
.0050	4.4602	.0050	1.3566				
.0100	4.9679	.0100	.5501	.0100	3.1885	.0100	.4495
.0175	5.4874	.0175	.2937				
.0500	5.1071	.0500	.0017	.0500	3.1592	.0500	.0121
.0750	4.3465	.0750	.0900				
.1000	4.1105	.1000	.1936	.1000	2.8469	.1000	.0672
.2000	3.4585	.2000	.4099				
.3000	3.1834	.3000	.4529	.3000	1.8771	.3000	.2480
.4000	2.8625	.4000	.4615				
.5000	2.6520	.5000	.4357	.5000	1.4465	.5000	.3634
.6000	2.4534	.6000	.4219				
.7000	2.2989	.7000	.3978	.7000	1.3501	.7000	.4942
.8000	2.1121	.8000	.3978				
.9000	1.9270	.9000	.2893	.9000	1.2192	.8792	.6372
1.0000	2.2004	1.0000	2.2004	1.0000	1.0505	1.0000	1.0505

ALPHA = 8.280 DEG DYNAMIC PRESSURE = 8.940 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	3.6894	0.0000	3.6894				
.0050	5.4564	.0050	1.7062				
.0100	5.8618	.0100	.7990	.0100	2.8531	.0100	.4164
.0175	5.6270	.0175	.2585				
.0500	5.1371	.0500	.0236	.0500	2.5544	.0500	.0103
.0750	4.6236	.0750	.0946				
.1000	4.3009	.1000	.1537	.1000	2.1356	.1000	.0565
.2000	3.5830	.2000	.3581				
.3000	3.2654	.3000	.4095	.3000	1.5969	.3000	.2450
.4000	2.8633	.4000	.4215				
.5000	2.5829	.5000	.4232	.5000	1.3879	.5000	.3718
.6000	2.4106	.6000	.4129				
.7000	2.2366	.7000	.3718	.7000	1.3433	.7000	.5226
.8000	2.0660	.8000	.3581				
.9000	1.8633	.9000	.2810	.9000	1.2799	.8792	.7008
1.0000	2.0728	1.0000	2.0728	1.0000	1.1274	1.0000	1.1274

ALPHA = 9.330 DEG DYNAMIC PRESSURE = 8.940 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	4.6219	0.0000	4.6219				
.0050	6.0291	.0050	2.1116				
.0100	6.1085	.0100	.9646	.0100	2.4669	.0100	.4009
.0175	5.8044	.0175	.3379				
.0500	5.4682	.0500	.0068	.0500	2.3913	.0500	0.0000
.0750	4.9040	.0750	.0574				
.1000	4.5087	.1000	.1284	.1000	2.2317	.1000	.0463
.2000	3.6708	.2000	.3016				
.3000	3.2772	.3000	.3752	.3000	1.4804	.3000	.0127
.4000	2.8819	.4000	.4044				
.5000	2.6302	.5000	.4146	.5000	1.3759	.5000	.3769
.6000	2.3599	.6000	.4009				
.7000	2.1302	.7000	.3838	.7000	1.3604	.7000	.5277
.8000	1.9477	.8000	.3615				
.9000	1.7957	.9000	.2998	.9000	1.3433	.8792	.7179
1.0000	1.8616	1.0000	1.8616	1.0000	1.2868	1.0000	1.2868

TABLE VI. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 10.320 DEG DYNAMIC PRESSURE = 8.940 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	4.4901	0.0000	4.4901				
.0050	6.4615	.0050	2.6252				
.0100	6.2183	.0100	1.2028	.0100	2.5047	.0100	.3769
.0175	5.9497	.0175	.5152				
.0500	5.3517	.0500	.0135	.0500	2.2952	.0500	.0291
.0750	4.9429	.0750	.0439				
.1000	4.5239	.1000	.1368	.1000	1.9862	.1000	.0565
.2000	3.6066	.2000	.3255				
.3000	3.2738	.3000	.3821	.3000	1.4667	.3000	.2673
.4000	2.8667	.4000	.4352				
.5000	2.5390	.5000	.4249	.5000	1.3947	.5000	.3838
.6000	2.2721	.6000	.3907				
.7000	2.0390	.7000	.3615	.7000	1.3707	.7000	.5363
.8000	1.8515	.8000	.3632				
.9000	1.7552	.9000	.3136	.9000	1.3159	.8792	.7265
1.0000	1.7197	1.0000	1.7197	1.0000	1.2645	1.0000	1.2645

ALPHA = 11.290 DEG DYNAMIC PRESSURE = 8.895 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	4.9746	0.0000	4.9746				
.0050	6.7319	.0050	2.7624				
.0100	6.5825	.0100	1.2853	.0100	2.4017	.0100	.3702
.0175	6.1699	.0175	.6112				
.0500	5.6181	.0500	.0051	.0500	2.2085	.0500	-.0086
.0750	4.9169	.0750	.0475				
.1000	4.5077	.1000	.0917	.1000	1.9756	.1000	.0551
.2000	3.7183	.2000	.3014				
.3000	3.2751	.3000	.3530	.3000	1.4724	.3000	.2497
.4000	2.7250	.4000	.3944				
.5000	2.5128	.5000	.4047	.5000	1.3880	.5000	.3789
.6000	2.2326	.6000	.3978				
.7000	1.9916	.7000	.3720	.7000	1.3604	.7000	.5321
.8000	1.8727	.8000	.3634				
.9000	1.6554	.9000	.3341	.9000	1.3312	.8792	.7233
1.0000	1.5620	1.0000	1.5620	1.0000	1.4517	1.0000	1.4517

ALPHA = 12.280 DEG DYNAMIC PRESSURE = 8.730 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	5.1067	0.0000	5.1067				
.0050	6.3990	.0050	2.8838				
.0100	6.7311	.0100	1.3684	.0100	2.9024	.0100	.3878
.0175	6.2831	.0175	.6020				
.0500	4.9839	.0500	.0259	.0500	2.7055	.0500	.0316
.0750	4.7383	.0750	.0536				
.1000	4.3092	.1000	.1124	.1000	2.5209	.1000	.0877
.2000	3.7626	.2000	.3071				
.3000	3.1796	.3000	.3878	.3000	1.5318	.3000	.2825
.4000	2.8353	.4000	.4018				
.5000	2.5707	.5000	.4036	.5000	1.4090	.5000	.4053
.6000	2.0794	.6000	.4334				
.7000	2.0448	.7000	.3808	.7000	1.4072	.7000	.5439
.8000	1.8147	.8000	.3878				
.9000	1.7784	.9000	.3422	.9000	1.3598	.8792	.7211
1.0000	1.6711	1.0000	1.6711	1.0000	1.3370	1.0000	1.3370

TABLE VI. - CONCLUDED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 13.240 DEG DYNAMIC PRESSURE = 8.748 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C'P	X/C	C'P	X/C	C'P	X/C	C'P
0.0000	5.1636	0.0000	5.1636				
.0050	6.7017	.0050	2.3444				
.0100	6.6810	.0100	1.2654	.0100	3.1122	.0100	.4500
.0175	6.0112	.0175	.4851				
.0500	5.5002	.0500	.0276	.0500	2.9965	.0500	.0228
.0750	4.6180	.0750	.0691				
.1000	4.2417	.1000	.1088	.1000	2.8333	.1000	.0735
.2000	3.5477	.2000	.3082				
.3000	3.1385	.3000	.3782	.3000	2.2438	.3000	.2802
.4000	2.6223	.4000	.4185				
.5000	2.3703	.5000	.4220	.5000	1.7650	.5000	.4027
.6000	2.2097	.6000	.4045				
.7000	2.1234	.7000	.4202	.7000	1.5934	.7000	.5971
.8000	2.0492	.8000	.4027				
.9000	2.0078	.9000	.3520	.9000	1.4288	.8792	.8440
1.0000	2.5775	1.0000	2.5775	1.0000	1.3150	1.0000	1.3150

ALPHA = 14.330 DEG DYNAMIC PRESSURE = 8.757 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C'P	X/C	C'P	X/C	C'P	X/C	C'P
0.0000	5.6170	0.0000	5.6170				
.0050	6.5483	.0050	2.3041				
.0100	6.5327	.0100	1.2676	.0100	3.1669	.0100	.4635
.0175	6.2741	.0175	.6364				
.0500	5.5687	.0500	.0190	.0500	3.0249	.0500	.0192
.0750	4.7806	.0750	.0431				
.1000	4.3184	.1000	.1397	.1000	2.5412	.1000	.0735
.2000	3.4112	.2000	.2764				
.3000	2.9525	.3000	.3761	.3000	1.8279	.3000	.2746
.4000	2.6007	.4000	.4128				
.5000	2.3420	.5000	.4303	.5000	1.7370	.5000	.4006
.6000	1.8919	.6000	.4251				
.7000	1.6556	.7000	.3866	.7000	1.7562	.7000	.5790
.8000	1.9867	.8000	.3656				
.9000	1.9505	.9000	.3551	.9000	1.4204	.8792	.7592
1.0000	2.6059	1.0000	2.6059	1.0000	1.3626	1.0000	1.3626

TABLE VII

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -4.610 DEG DYNAMIC PRESSURE = 21.402 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	-.0014	0.0000	-.0014				
.0050	.3168	.0050	.1630				
.0100	.7692	.0100	.5151	.0100	3.2778	.0100	.5504
.0175	.8743	.0175	.7635				
.0500	1.7444	.0500	1.0782	.0500	3.2362	.0500	.0072
.0750	1.8403	.0750	1.0923				
.1000	2.0153	.1000	1.0916	.1000	3.1939	.1000	.0551
.2000	2.1226	.2000	.9179				
.3000	2.2397	.3000	.8311	.3000	2.2739	.3000	.2863
.4000	2.1748	.4000	.7329				
.5000	2.1155	.5000	.6671	.5000	1.5418	.5000	.4158
.6000	2.0781	.6000	.6034				
.7000	2.0633	.7000	.5253	.7000	1.2183	.7000	.5482
.8000	2.0139	.8000	.4387				
.9000	1.9144	.9000	.3750	.9000	1.1983	.8792	.7071
1.0000	2.4274	1.0000	2.4274	1.0000	1.1137	1.0000	1.1137

ALPHA = -3.580 DEG DYNAMIC PRESSURE = 21.384 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	.0078	0.0000	.0078				
.0050	.5085	.0050	.0975				
.0100	.9619	.0100	.3100	.0100	3.3265	.0100	.5286
.0175	1.2034	.0175	.5170				
.0500	2.0022	.0500	.9570	.0500	3.2914	.0500	.0007
.0750	2.1074	.0750	.9937				
.1000	2.1145	.1000	1.0022	.1000	3.2483	.1000	.0580
.2000	2.2670	.2000	.8605				
.3000	2.2917	.3000	.8088	.3000	2.2923	.3000	.2865
.4000	2.2360	.4000	.7213				
.5000	2.1957	.5000	.6332	.5000	1.5531	.5000	.4133
.6000	2.1406	.6000	.5752				
.7000	2.1004	.7000	.4993	.7000	1.2186	.7000	.5458
.8000	2.0481	.8000	.4327				
.9000	1.9026	.9000	.3539	.9000	1.1785	.8792	.7006
1.0000	2.4577	1.0000	2.4577	1.0000	1.1060	1.0000	1.1060

ALPHA = -2.600 DEG DYNAMIC PRESSURE = 21.356 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	.0205	0.0000	.0205				
.0050	.8585	.0050	.0014				
.0100	1.4348	.0100	.1464	.0100	3.3510	.0100	.4813
.0175	1.4985	.0175	.2708				
.0500	2.1795	.0500	.7588	.0500	3.3064	.0500	.0057
.0750	2.3209	.0750	.8868				
.1000	2.2948	.1000	.9137	.1000	3.2245	.1000	.0653
.2000	2.3287	.2000	.8315				
.3000	2.4178	.3000	.7352	.3000	2.2184	.3000	.2819
.4000	2.3372	.4000	.6742				
.5000	2.2587	.5000	.6226	.5000	1.4962	.5000	.4096
.6000	2.1554	.6000	.5645				
.7000	2.1385	.7000	.4849	.7000	1.1944	.7000	.5415
.8000	2.0501	.8000	.4210				
.9000	1.9256	.9000	.3486	.9000	1.1628	.8792	.6864
1.0000	2.4553	1.0000	2.4553	1.0000	1.0916	1.0000	1.0916

Note: The conversion factor from pounds force/square foot to newtons/square meter is 47.880258.

TABLE VII. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = -1.550 DEG DYNAMIC PRESSURE = 21.420 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	.1622	0.0000	.1622				
.0050	.9843	.0050	-.0014				
.0100	1.4489	.0100	.0571	.0100	3.3439	.0100	.4691
.0175	1.8204	.0175	.1417				
.0500	2.4818	.0500	.6902	.0500	3.2808	.0500	-.0021
.0750	2.4423	.0750	.7960				
.1000	2.5375	.1000	.7509	.1000	3.2457	.1000	.0672
.2000	2.5170	.2000	.7516				
.3000	2.5206	.3000	.7058	.3000	2.1767	.3000	.2725
.4000	2.3774	.4000	.6615				
.5000	2.2942	.5000	.6164	.5000	1.3277	.5000	.4098
.6000	2.2202	.6000	.5320				
.7000	2.1427	.7000	.4770	.7000	1.2023	.7000	.5270
.8000	2.0559	.8000	.3990				
.9000	1.9368	.9000	.3282	.9000	1.1944	.8792	.6822
1.0000	2.4437	1.0000	2.4437	1.0000	1.1227	1.0000	1.1227

ALPHA = -.570 DEG DYNAMIC PRESSURE = 21.439 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	.3804	0.0000	.3804				
.0050	1.5674	.0050	-.0021				
.0100	1.9703	.0100	-.0070	.0100	3.3688	.0100	.4494
.0175	1.9794	.0175	.0986				
.0500	2.6332	.0500	.6129	.0500	3.3201	.0500	-.0021
.0750	2.6198	.0750	.7319				
.1000	2.7233	.1000	.7037	.1000	3.2214	.1000	.0593
.2000	2.5852	.2000	.6845				
.3000	2.6198	.3000	.6452	.3000	2.2027	.3000	.2758
.4000	2.4951	.4000	.6209				
.5000	2.3598	.5000	.6009	.5000	1.3723	.5000	.3958
.6000	2.2373	.6000	.5287				
.7000	2.1725	.7000	.4608	.7000	1.2377	.7000	.5209
.8000	2.0964	.8000	.3958				
.9000	1.9668	.9000	.3294	.9000	1.1869	.8792	.6680
1.0000	2.4486	1.0000	2.4486	1.0000	1.0738	1.0000	1.0738

ALPHA = .470 DEG DYNAMIC PRESSURE = 21.256 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	.8874	0.0000	.8874				
.0050	1.9837	.0050	.0313				
.0100	2.2018	.0100	-.0078	.0100	3.3437	.0100	.4396
.0175	2.4334	.0175	.0355				
.0500	2.8320	.0500	.4178	.0500	3.2888	.0500	-.0036
.0750	2.8597	.0750	.5656				
.1000	2.8889	.1000	.6110	.1000	3.2310	.1000	.0620
.2000	2.7709	.2000	.6032				
.3000	2.6381	.3000	.6053	.3000	2.1834	.3000	.2645
.4000	2.5045	.4000	.5960				
.5000	2.3659	.5000	.5571	.5000	1.3437	.5000	.3891
.6000	2.2395	.6000	.5275				
.7000	2.2075	.7000	.4446	.7000	1.2094	.7000	.5124
.8000	2.1137	.8000	.3798				
.9000	1.9801	.9000	.3084	.9000	1.1812	.8792	.6724
1.0000	2.4754	1.0000	2.4754	1.0000	1.0773	1.0000	1.0773

TABLE VII. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 1.500 DEG DYNAMIC PRESSURE = 21.393 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	1.0794	0.0000	1.0794				
.0050	2.2703	.0050	.2012				
.0100	2.5809	.0100	.0014	.0100	3.3345	.0100	.4411
.0175	2.9367	.0175	.0134				
.0500	3.5227	.0500	.3092	.0500	3.2613	.0500	-.0050
.0750	3.0624	.0750	.4299				
.1000	2.9805	.1000	.5408	.1000	3.1522	.1000	.0601
.2000	2.8428	.2000	.5671				
.3000	2.7948	.3000	.5814	.3000	2.1780	.3000	.2606
.4000	2.5739	.4000	.5778				
.5000	2.4291	.5000	.5248	.5000	1.3630	.5000	.3773
.6000	2.3127	.6000	.4933				
.7000	2.2442	.7000	.4368	.7000	1.2224	.7000	.5041
.8000	2.1355	.8000	.3759				
.9000	2.0035	.9000	.3007	.9000	1.1815	.8792	.6480
1.0000	2.4320	1.0000	2.4320	1.0000	1.0739	1.0000	1.0739

ALPHA = 2.550 DEG DYNAMIC PRESSURE = 21.439 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	1.2320	0.0000	1.2320				
.0050	2.7945	.0050	.3346				
.0100	3.0650	.0100	.0282	.0100	3.3044	.0100	.4294
.0175	3.4728	.0175	-.0014				
.0500	3.4961	.0500	.2029	.0500	3.2543	.0500	.0036
.0750	3.2580	.0750	.3564				
.1000	3.2291	.1000	.4501	.1000	3.1369	.1000	.0622
.2000	2.9227	.2000	.5273				
.3000	2.8487	.3000	.5394	.3000	2.1447	.3000	.2551
.4000	2.5775	.4000	.5344				
.5000	2.4331	.5000	.4994	.5000	1.3150	.5000	.3722
.6000	2.3331	.6000	.4916				
.7000	2.2492	.7000	.4480	.7000	1.1840	.7000	.4951
.8000	2.1316	.8000	.3601				
.9000	2.0104	.9000	.2987	.9000	1.1754	.8792	.6488
1.0000	2.4289	1.0000	2.4289	1.0000	1.0910	1.0000	1.0910

ALPHA = 3.560 DEG DYNAMIC PRESSURE = 21.356 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	CP	X/C	CP	X/C	CP	X/C	CP
0.0000	1.6046	0.0000	1.6046				
.0050	3.1462	.0050	.4604				
.0100	3.5535	.0100	.0834	.0100	3.3007	.0100	.4311
.0175	3.5280	.0175	.0007				
.0500	3.7692	.0500	.1832	.0500	3.2411	.0500	.0022
.0750	3.6334	.0750	.3437				
.1000	3.5273	.1000	.3861	.1000	3.1009	.1000	.0588
.2000	3.1370	.2000	.4999				
.3000	2.9560	.3000	.5179	.3000	2.1207	.3000	.2444
.4000	2.6604	.4000	.5093				
.5000	2.4963	.5000	.4913	.5000	1.3568	.5000	.3751
.6000	2.3761	.6000	.4734				
.7000	2.2969	.7000	.4347	.7000	1.1412	.7000	.4834
.8000	2.1604	.8000	.3493				
.9000	2.0098	.9000	.2812	.9000	1.1441	.8792	.6484
1.0000	2.4093	1.0000	2.4093	1.0000	1.0701	1.0000	1.0701

TABLE VII. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 4.590 DEG DYNAMIC PRESSURE = 21.384 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P	X/C	C_P	X/C	C_P	X/C	C_P
0.0000	1.7712	0.0000	1.7712				
.0050	3.6950	.0050	.4831				
.0100	4.0750	.0100	.1787	.0100	3.2820	.0100	.4255
.0175	3.9754	.0175	.0438				
.0500	4.2353	.0500	.1073	.0500	3.1880	.0500	-.0050
.0750	3.8123	.0750	.2260				
.1000	3.6929	.1000	.3199	.1000	3.0588	.1000	.0559
.2000	3.1682	.2000	.4835				
.3000	3.0058	.3000	.4721	.3000	2.0993	.3000	.2414
.4000	2.7275	.4000	.4864				
.5000	2.5615	.5000	.4678	.5000	1.3550	.5000	.3574
.6000	2.4076	.6000	.4434				
.7000	2.2995	.7000	.4133	.7000	1.1469	.7000	.4778
.8000	2.1738	.8000	.3395				
.9000	2.0135	.9000	.2794	.9000	1.0780	.8792	.6246
1.0000	2.3909	1.0000	2.3909	1.0000	1.0737	1.0000	1.0737

ALPHA = 5.650 DEG DYNAMIC PRESSURE = 21.457 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P	X/C	C_P	X/C	C_P	X/C	C_P
0.0000	2.4036	0.0000	2.4036				
.0050	3.8993	.0050	.9896				
.0100	4.1942	.0100	.4448	.0100	3.2408	.0100	.4112
.0175	4.5355	.0175	.0957				
.0500	4.7650	.0500	.0831	.0500	3.1922	.0500	.0021
.0750	4.0231	.0750	.1964				
.1000	3.8725	.1000	.2766	.1000	3.0634	.1000	.0564
.2000	3.3657	.2000	.4269				
.3000	3.0603	.3000	.4712	.3000	2.1479	.3000	.2399
.4000	2.7872	.4000	.4762				
.5000	2.5894	.5000	.4483	.5000	1.4141	.5000	.3569
.6000	2.4339	.6000	.4169				
.7000	2.3269	.7000	.4112	.7000	1.1415	.7000	.4676
.8000	2.1882	.8000	.3427				
.9000	2.0313	.9000	.2684	.9000	1.0579	.8792	.6054
1.0000	2.3677	1.0000	2.3677	1.0000	1.0335	1.0000	1.0335

ALPHA = 6.650 DEG DYNAMIC PRESSURE = 21.530 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P	X/C	C_P	X/C	C_P	X/C	C_P
0.0000	3.6482	0.0000	3.6482				
.0050	5.3352	.0050	1.3762				
.0100	5.0476	.0100	.4658	.0100	3.0509	.0100	.3757
.0175	4.6857	.0175	.1494				
.0500	4.6520	.0500	.0533	.0500	3.1721	.0500	.0028
.0750	4.2606	.0750	.1662				
.1000	4.0418	.1000	.2441	.1000	3.0445	.1000	.0591
.2000	3.4750	.2000	.3401				
.3000	3.1369	.3000	.4006	.3000	2.0922	.3000	.2391
.4000	2.7855	.4000	.4326				
.5000	2.6143	.5000	.4376	.5000	1.4877	.5000	.3515
.6000	2.4663	.6000	.4198				
.7000	2.3246	.7000	.3842	.7000	1.2389	.7000	.4560
.8000	2.1338	.8000	.3486				
.9000	1.9472	.9000	.2632	.9000	1.1619	.8792	.6005
1.0000	2.1605	1.0000	2.1605	1.0000	1.0386	1.0000	1.0386

TABLE VII. - CONTINUED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 7.690 DEG DYNAMIC PRESSURE = 21.493 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P	X/C	C_P	X/C	C_P	X/C	C_P
0.0000	3.6721	0.0000	3.6721				
.0050	5.0999	.0050	1.5388				
.0100	5.2749	.0100	.7034	.0100	2.8898	.0100	.3670
.0175	5.4456	.0175	.1736				
.0500	5.2489	.0500	.0232	.0500	2.7884	.0500	0.0000
.0750	4.5188	.0750	.1258				
.1000	4.1970	.1000	.2066	.1000	2.7670	.1000	.0577
.2000	3.5414	.2000	.3514				
.3000	3.2147	.3000	.4134	.3000	1.8308	.3000	.2395
.4000	2.8387	.4000	.4319				
.5000	2.6244	.5000	.4162	.5000	1.3367	.5000	.3528
.6000	2.4375	.6000	.4005				
.7000	2.2478	.7000	.3799	.7000	1.3039	.7000	.4789
.8000	2.0742	.8000	.3599				
.9000	1.8937	.9000	.2566	.9000	1.2189	.8792	.6393
1.0000	2.1846	1.0000	2.1846	1.0000	1.0304	1.0000	1.0304

ALPHA = 8.750 DEG DYNAMIC PRESSURE = 21.411 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P	X/C	C_P	X/C	C_P	X/C	C_P
0.0000	4.2039	0.0000	4.2039				
.0050	5.8036	.0050	1.8099				
.0100	5.6823	.0100	.8775	.0100	2.7260	.0100	.3570
.0175	5.3804	.0175	.3499				
.0500	5.2838	.0500	.0071	.0500	2.6206	.0500	.0057
.0750	4.7244	.0750	.0931				
.1000	4.3823	.1000	.1418	.1000	2.2307	.1000	.0630
.2000	3.6100	.2000	.3176				
.3000	3.2305	.3000	.3820	.3000	1.5318	.3000	.2490
.4000	2.8757	.4000	.4106				
.5000	2.6154	.5000	.4135	.5000	1.3483	.5000	.3677
.6000	2.4095	.6000	.3992				
.7000	2.2049	.7000	.3634	.7000	1.3304	.7000	.5058
.8000	2.0624	.8000	.3513				
.9000	1.9454	.9000	.2504	.9000	1.2286	.8792	.6947
1.0000	2.0497	1.0000	2.0497	1.0000	1.1340	1.0000	1.1340

ALPHA = 9.740 DEG DYNAMIC PRESSURE = 21.439 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P	X/C	C_P	X/C	C_P	X/C	C_P
0.0000	4.6682	0.0000	4.6682				
.0050	6.2342	.0050	2.5810				
.0100	6.1278	.0100	1.2468	.0100	2.4461	.0100	.3444
.0175	6.0447	.0175	.4889				
.0500	5.5720	.0500	.0035	.0500	2.3194	.0500	.0064
.0750	4.8408	.0750	.0740				
.1000	4.4872	.1000	.1395	.1000	2.1297	.1000	.0722
.2000	3.6757	.2000	.2972				
.3000	3.2798	.3000	.3701	.3000	1.4510	.3000	.2551
.4000	2.8783	.4000	.4001				
.5000	2.6177	.5000	.3944	.5000	1.3322	.5000	.3687
.6000	2.3472	.6000	.3894				
.7000	2.0809	.7000	.3622	.7000	1.3487	.7000	.5187
.8000	1.9625	.8000	.3430				
.9000	1.8949	.9000	.2494	.9000	1.2778	.8792	.6966
1.0000	1.8498	1.0000	1.8498	1.0000	1.2542	1.0000	1.2542

TABLE VII. - CONCLUDED

PRESSURE DISTRIBUTION ON 4415 WING

ALPHA = 10.680 DEG DYNAMIC PRESSURE = 21.439 LBF/SQ.FT. FLAP = 40 DEG

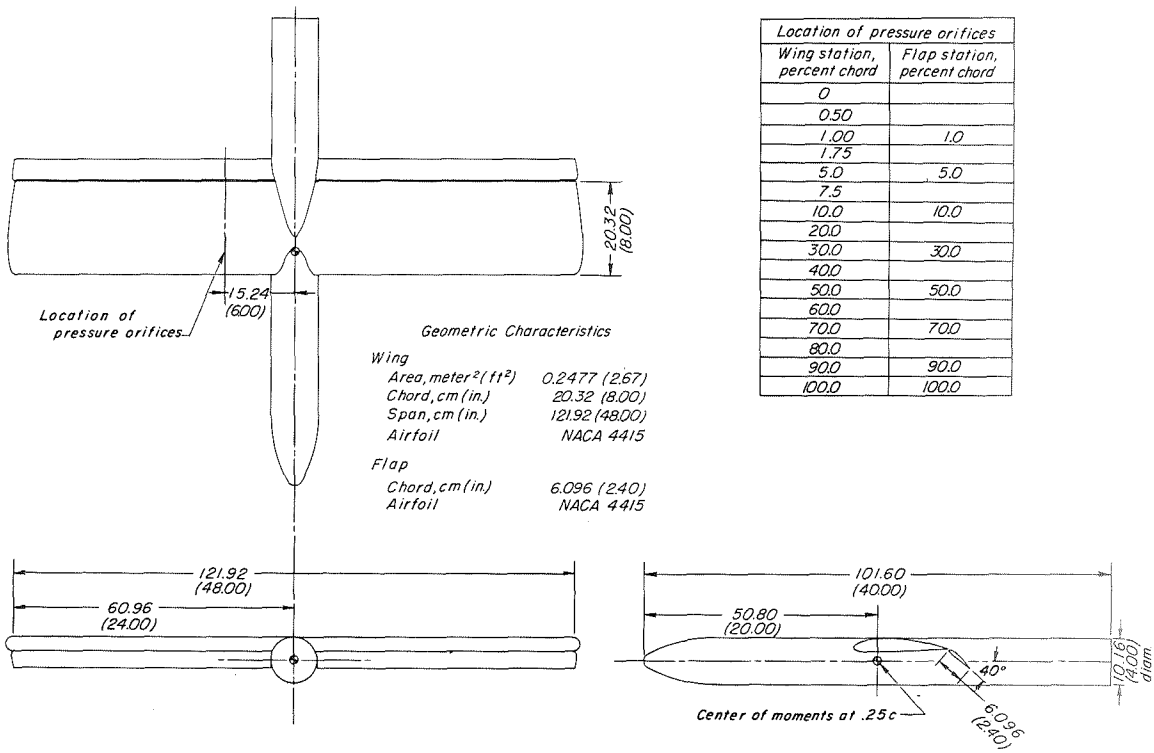
WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	5.1346	0.0000	5.1346				
.0050	6.3208	.0050	2.2612				
.0100		.0100	1.1081	.0100	2.5041	.0100	.3279
.0175		.0175	.5586				
.0500	5.5227	.0500	-.0049	.0500	2.2864	.0500	-.0021
.0750	4.8542	.0750	.0444				
.1000	4.5738	.1000	.1219	.1000	2.2106	.1000	.0636
.2000	3.6109	.2000	.2779				
.3000	3.2249	.3000	.3572	.3000	1.4560	.3000	.2508
.4000	2.8149	.4000	.3865				
.5000	2.5888	.5000	.3851	.5000	1.3630	.5000	.3722
.6000	2.3500	.6000	.3980				
.7000	2.2175	.7000	.3680	.7000	1.3444	.7000	.5066
.8000	1.9463	.8000	.3465				
.9000	1.7385	.9000	.2529	.9000	1.2842	.8792	.6952
1.0000	1.7449	1.0000	1.7449	1.0000	1.2184	1.0000	1.2184

ALPHA = 11.690 DEG DYNAMIC PRESSURE = 21.366 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	5.3741	0.0000	5.3741				
.0050		.0050	2.5976				
.0100		.0100	1.2999	.0100	2.4537	.0100	.3319
.0175	6.3544	.0175	.6128				
.0500	5.6250	.0500	-.0014	.0500	2.2217	.0500	.0022
.0750	4.8503	.0750	.0290				
.1000	4.4806	.1000	.1004	.1000	1.9631	.1000	.0602
.2000	3.6112	.2000	.2610				
.3000	3.2175	.3000	.3499	.3000	1.5501	.3000	.2495
.4000	2.7920	.4000	.3785				
.5000	2.4859	.5000	.3857	.5000	1.4036	.5000	.3800
.6000	2.3707	.6000	.3807				
.7000	2.0689	.7000	.3570	.7000	1.3641	.7000	.5097
.8000	1.9671	.8000	.3427				
.9000	1.8258	.9000	.2488	.9000	1.2836	.8792	.6990
1.0000	1.7035	1.0000	1.7035	1.0000	1.2477	1.0000	1.2477

ALPHA = 12.730 DEG DYNAMIC PRESSURE = 21.274 LBF/SQ.FT. FLAP = 40 DEG

WING UPPER SURFACE		WING LOWER SURFACE		FLAP UPPER SURFACE		FLAP LOWER SURFACE	
X/C	C_P'	X/C	C_P'	X/C	C_P'	X/C	C_P'
0.0000	5.8289	0.0000	5.8289				
.0050	6.0476	.0050	2.8687				
.0100		.0100	1.4886	.0100	2.3395	.0100	.3247
.0175		.0175	.6318				
.0500	5.6699	.0500	-.0007	.0500	2.4239	.0500	.0079
.0750	4.8890	.0750	.0390				
.1000	4.5575	.1000	.1036	.1000	1.9406	.1000	.0720
.2000	3.6084	.2000	.2419				
.3000	2.8956	.3000	.3701	.3000	1.5207	.3000	.2484
.4000	2.7863	.4000	.3859				
.5000	2.3682	.5000	.3795	.5000	1.4356	.5000	.3838
.6000	2.2078	.6000	.3903				
.7000	1.8848	.7000	.3600	.7000	1.4883	.7000	.5364
.8000	1.6824	.8000	.3391				
.9000	1.6271	.9000	.2635	.9000	1.5308	.8792	.7107
1.0000	1.6611	1.0000	1.6611	1.0000	1.3245	1.0000	1.3245



Location of pressure orifices	
Wing station, percent chord	Flap station, percent chord
0	
0.50	
1.00	1.0
1.75	
5.0	5.0
7.5	
10.0	10.0
20.0	
30.0	30.0
40.0	
50.0	50.0
60.0	
70.0	70.0
80.0	
90.0	90.0
100.0	100.0

Figure 1.- Three-view drawing of model. Dimensions are given in centimeters and parenthetically in inches unless otherwise noted.

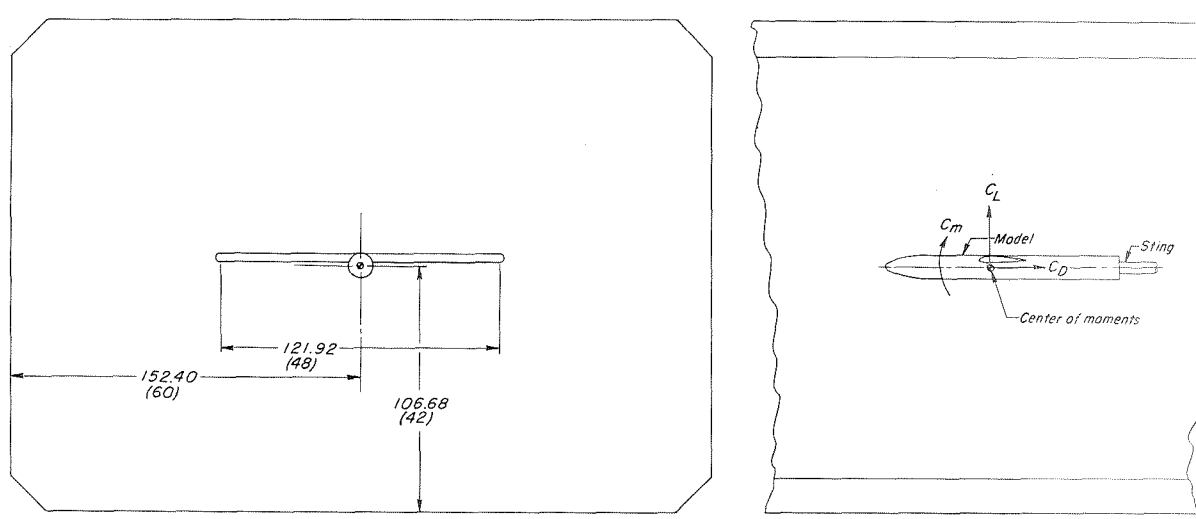


Figure 2.- Sketch showing setup of model in test section of the Langley 300-MPH 7- by 10-foot tunnel. Dimensions are given in centimeters and parenthetically in inches.

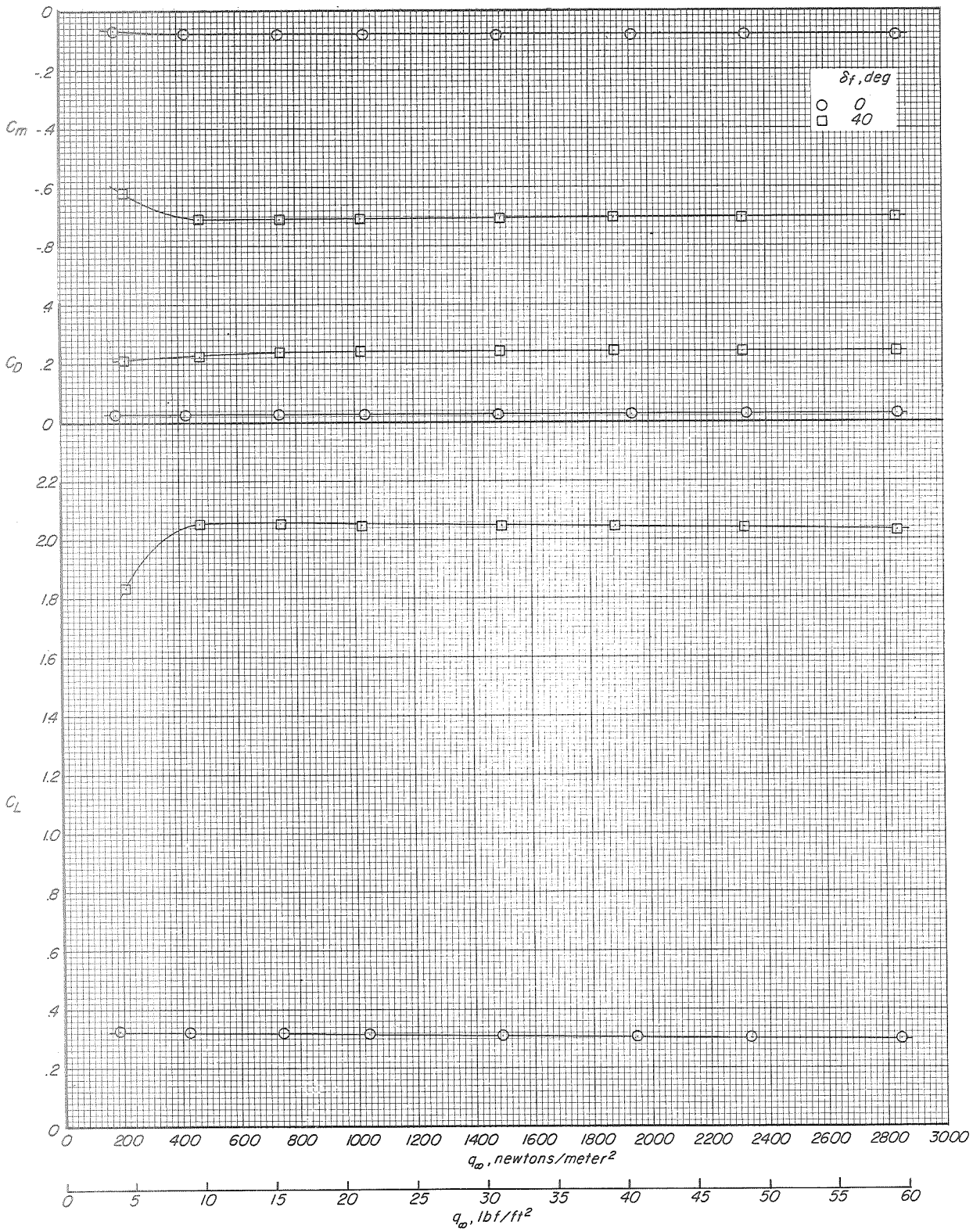


Figure 3.- Effect of variation in free-stream dynamic pressure on the longitudinal aerodynamic characteristics of the model. $\alpha \approx 0^\circ$.

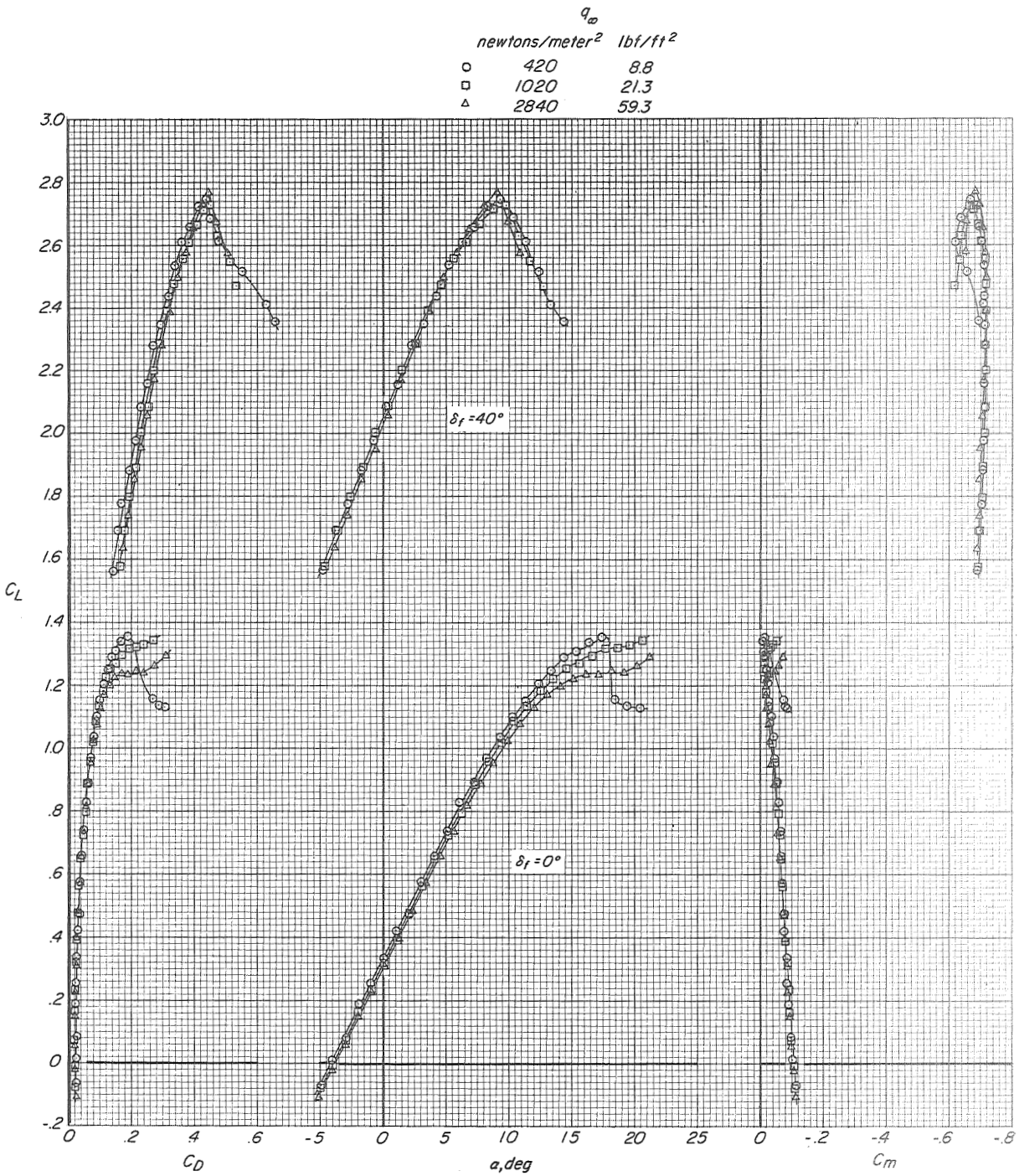


Figure 4.- Effect of angle of attack on the longitudinal aerodynamic characteristics of the model.

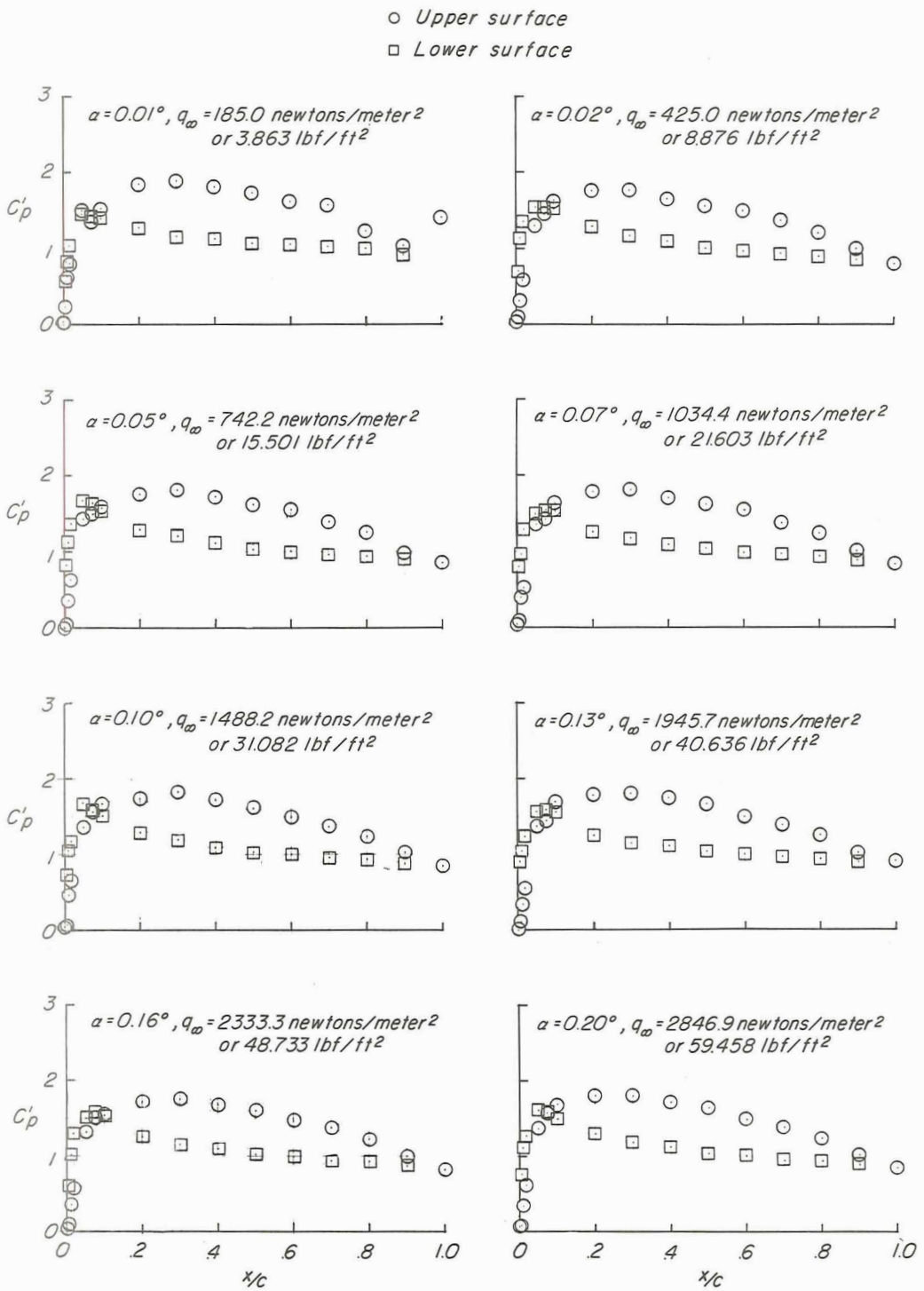
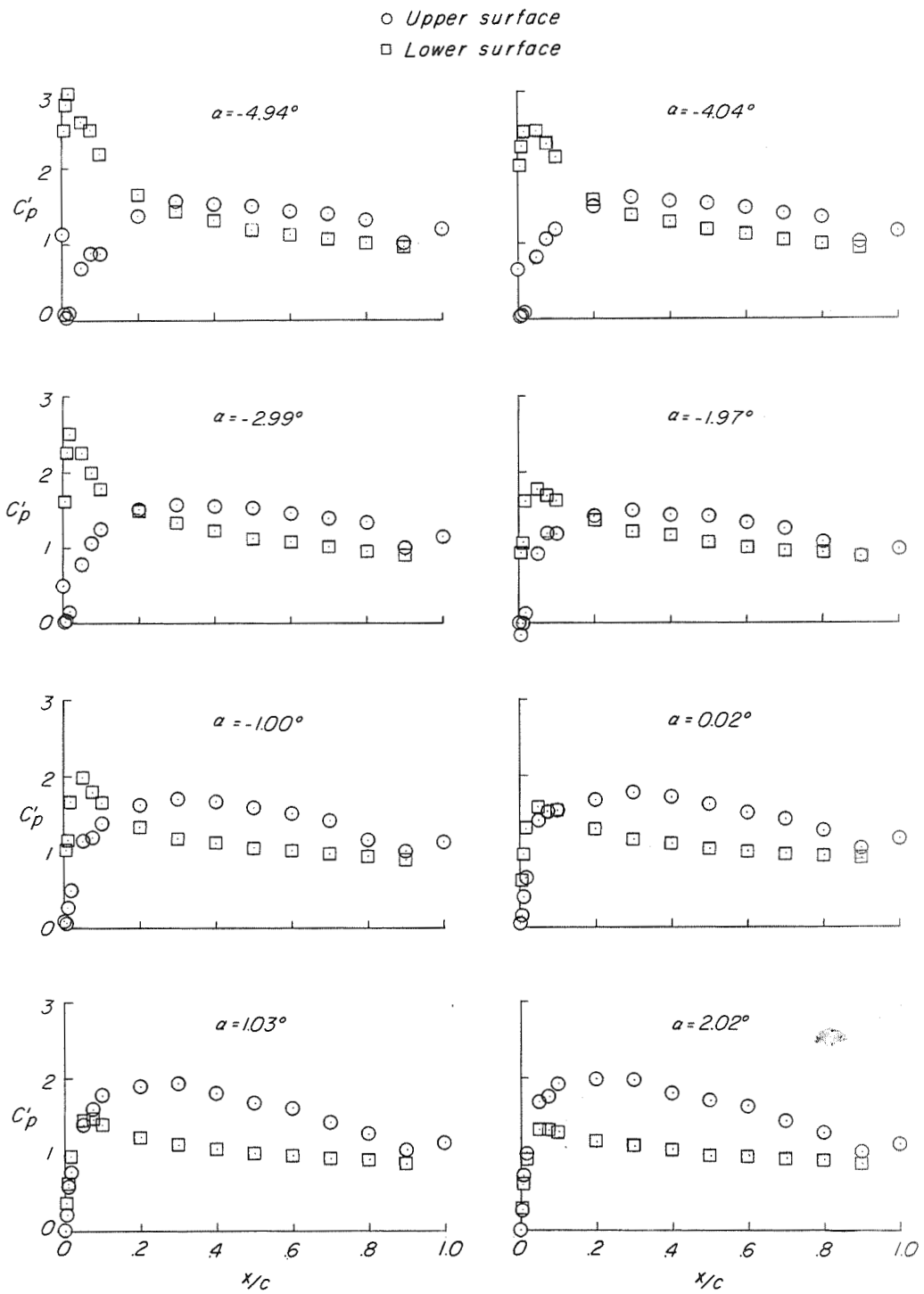
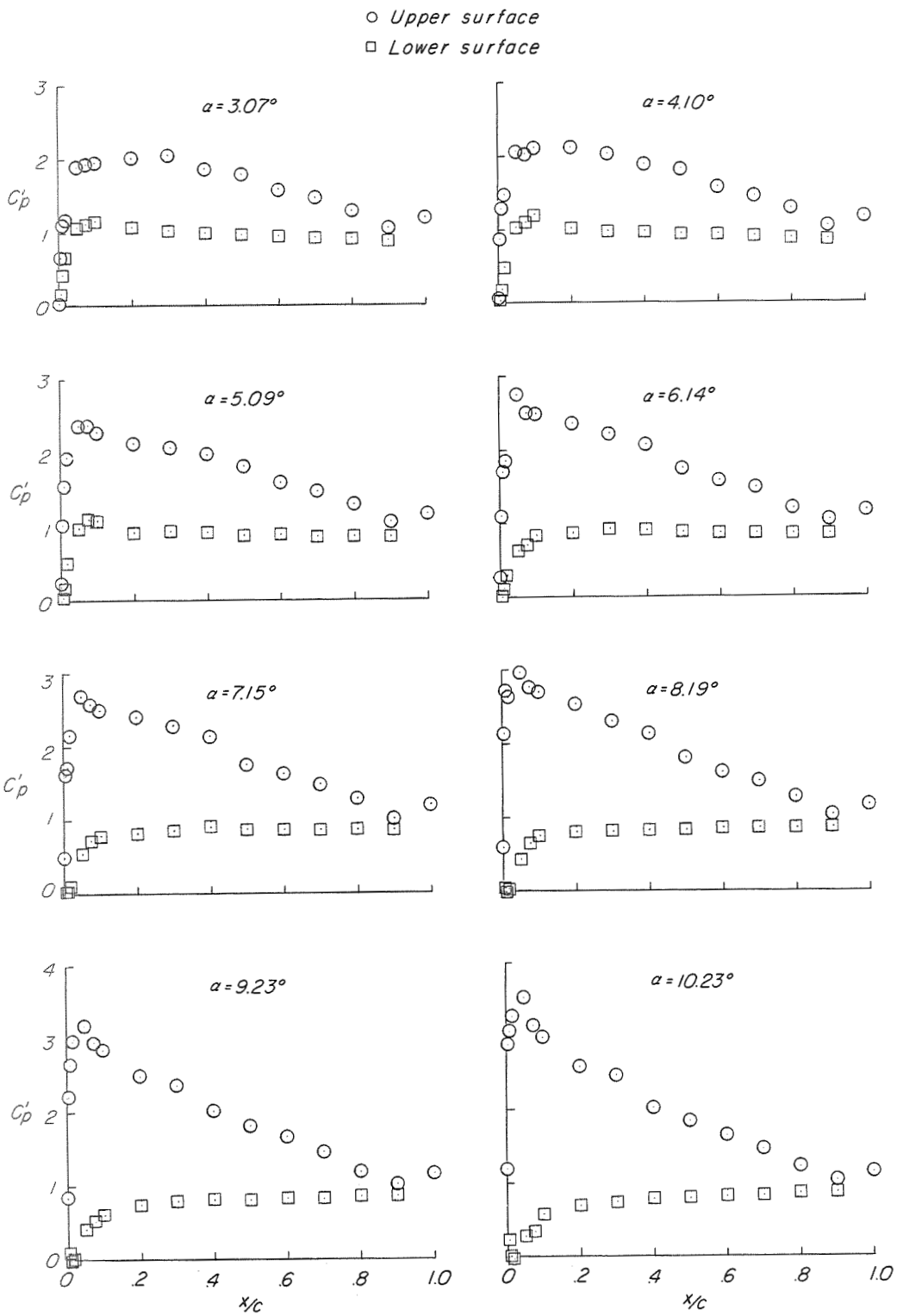


Figure 5.- Pressure distributions on wing of model at several free-stream dynamic pressures. $\delta_f \approx 0^o$.



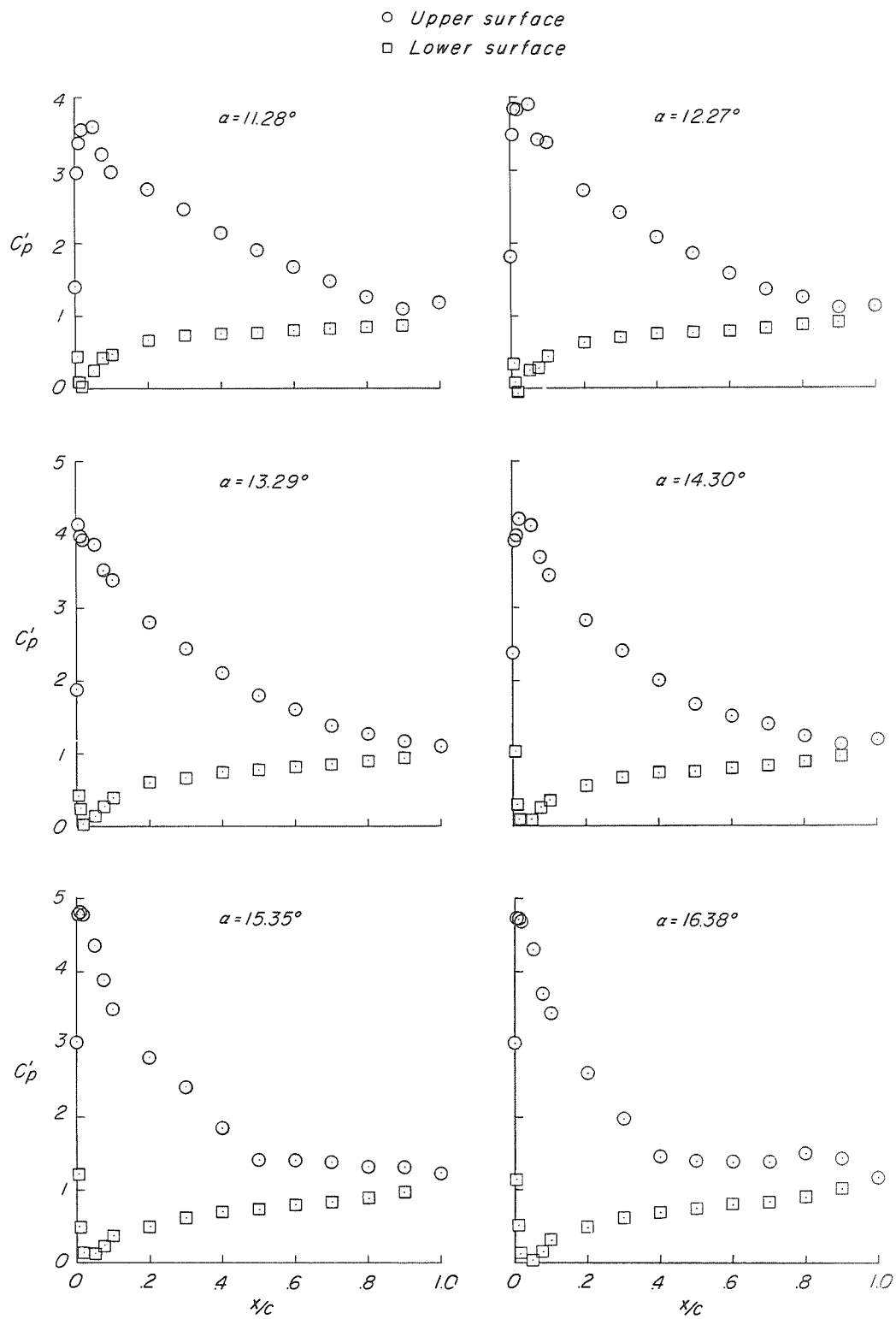
(a) $\alpha = -4.94^{\circ}$ to 2.02°.

Figure 6.- Pressure distributions on wing of model at free-stream dynamic pressure of approximately 420 N/m^2 (8.8 lbf/ft^2). $\delta_f = 0^{\circ}$.



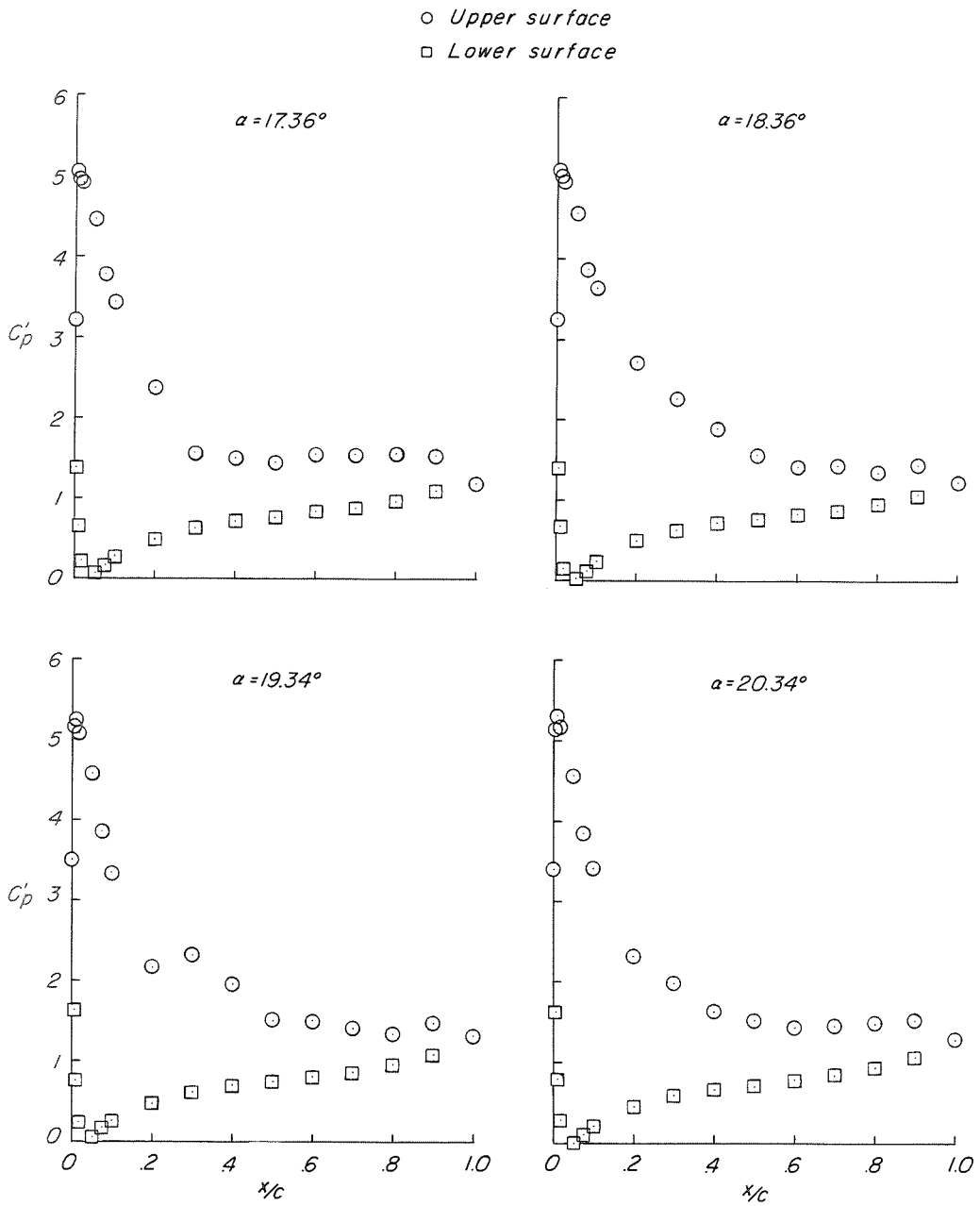
(b) $\alpha = 3.07^{\circ}$ to 10.23°.

Figure 6.- Continued.



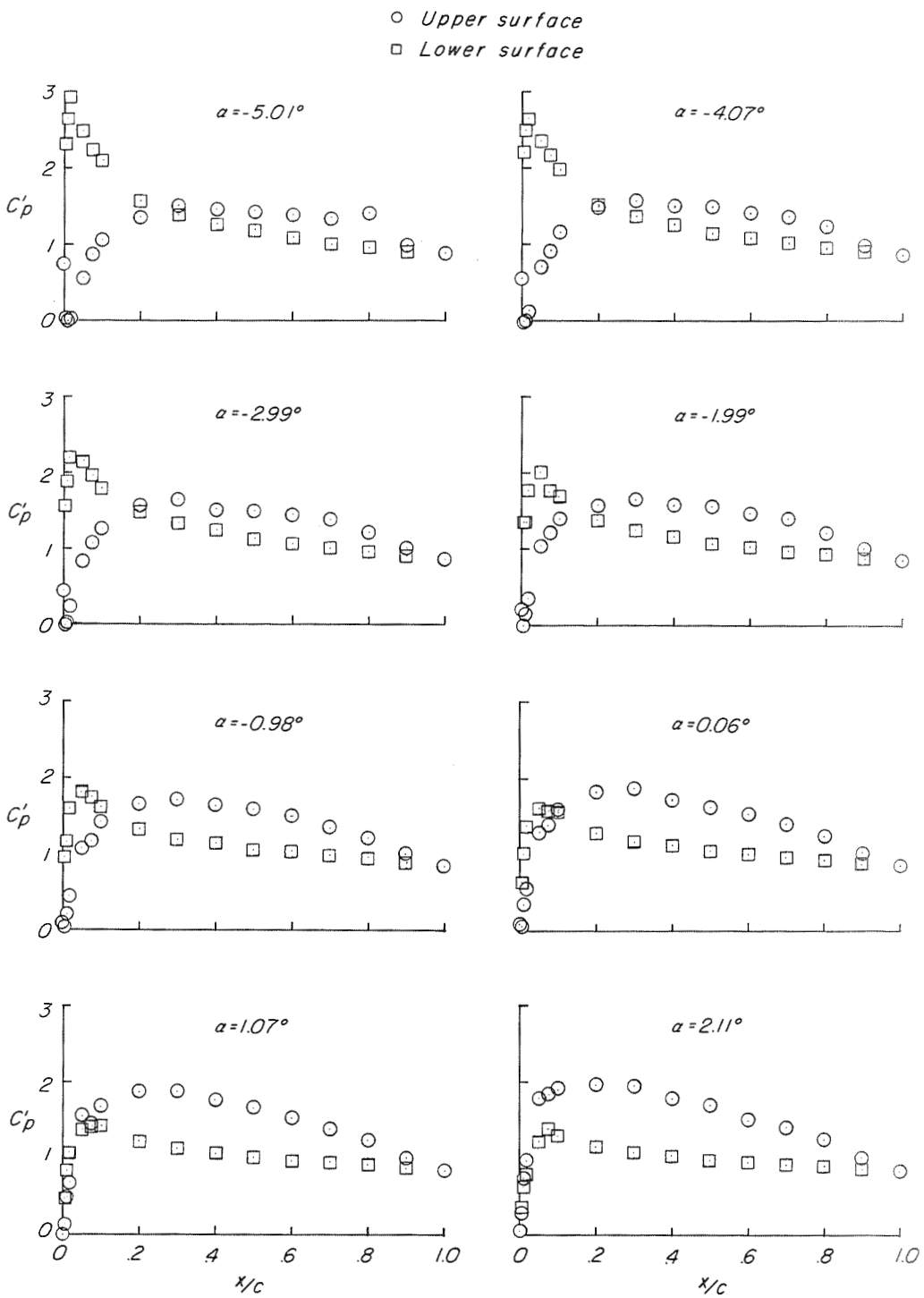
(c) $\alpha = 11.28^{\circ}$ to 16.38°.

Figure 6.- Continued.



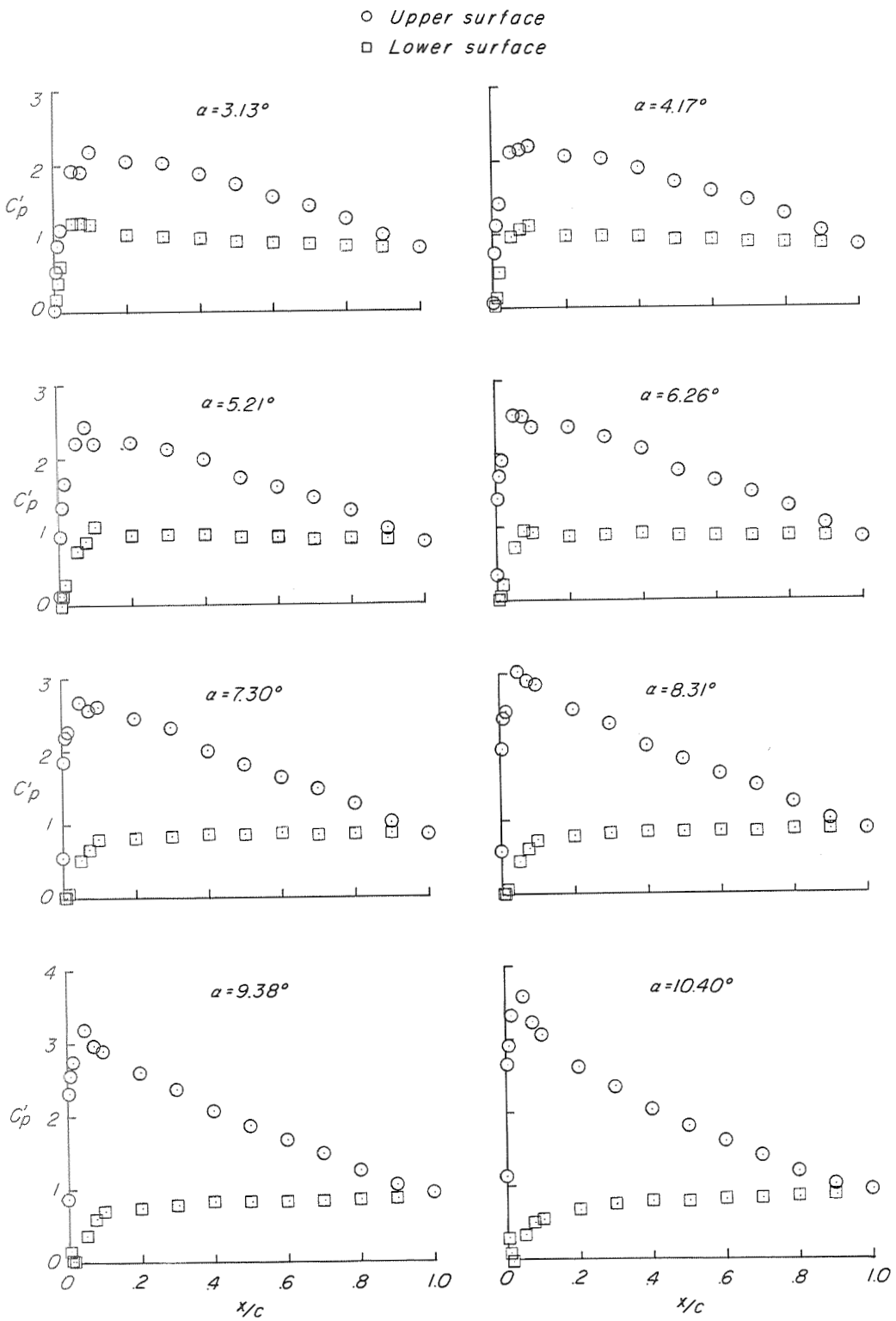
(d) $\alpha = 17.36^{\circ}$ to 20.34°.

Figure 6.- Concluded.



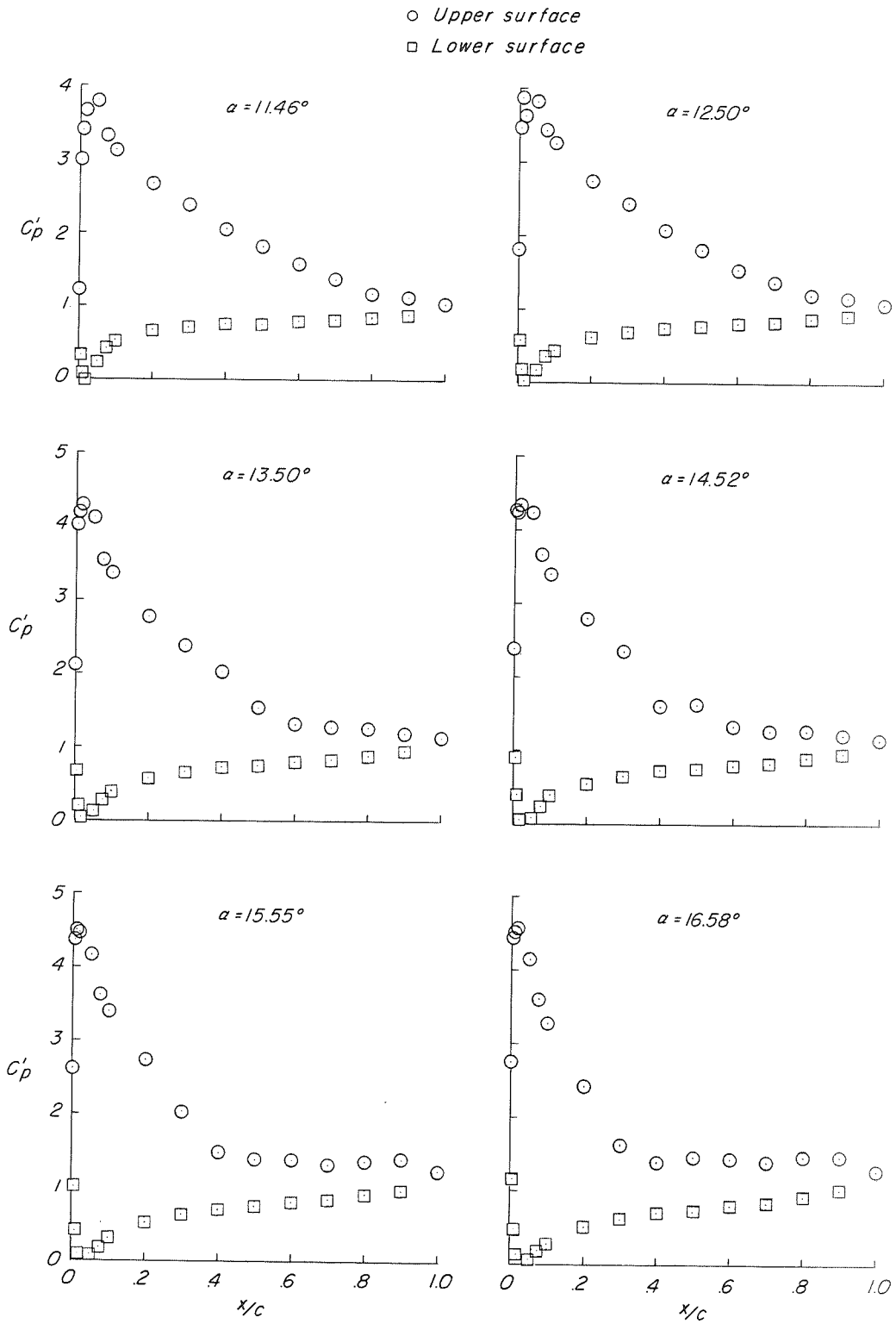
(a) α = -5.01° to 2.11°.

Figure 7.- Pressure distributions on wing of model at free-stream dynamic pressure of approximately 1020 N/m^2 (21.3 lbf/ft^2). δ_f = 0°.



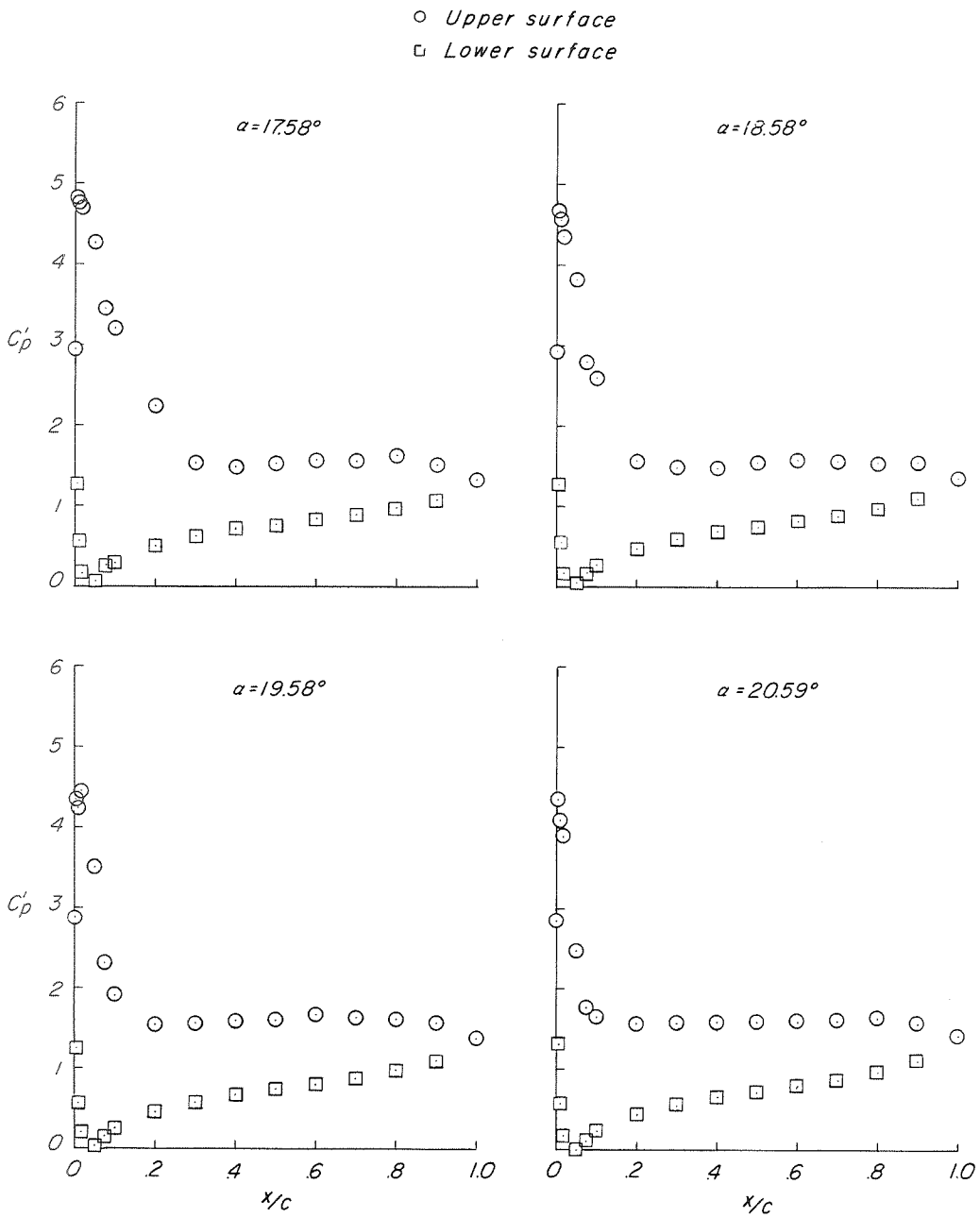
(b) $\alpha = 3.13^{\circ}$ to 10.40°.

Figure 7.- Continued.



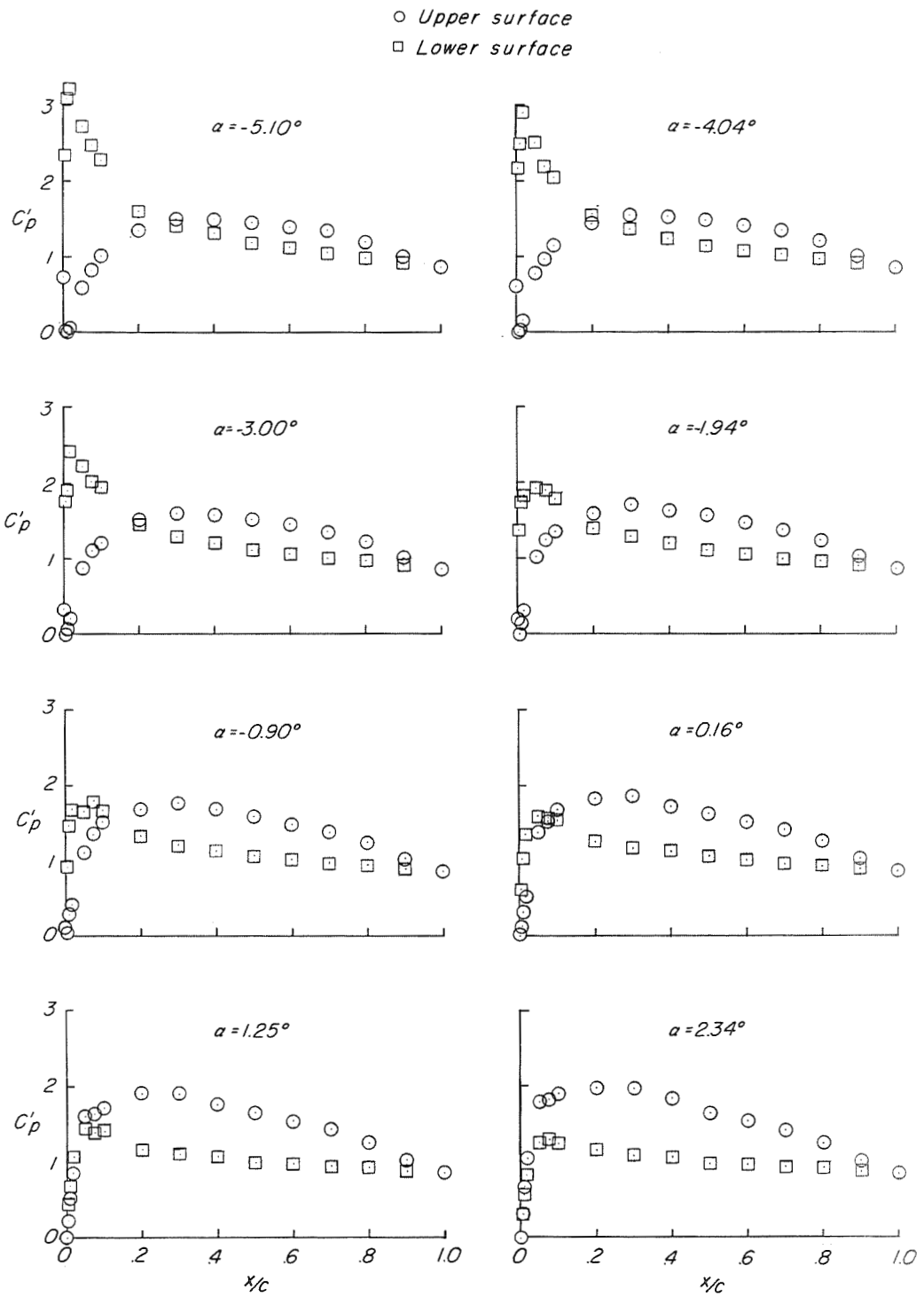
(c) α = 11.46° to 16.58°.

Figure 7.- Continued.



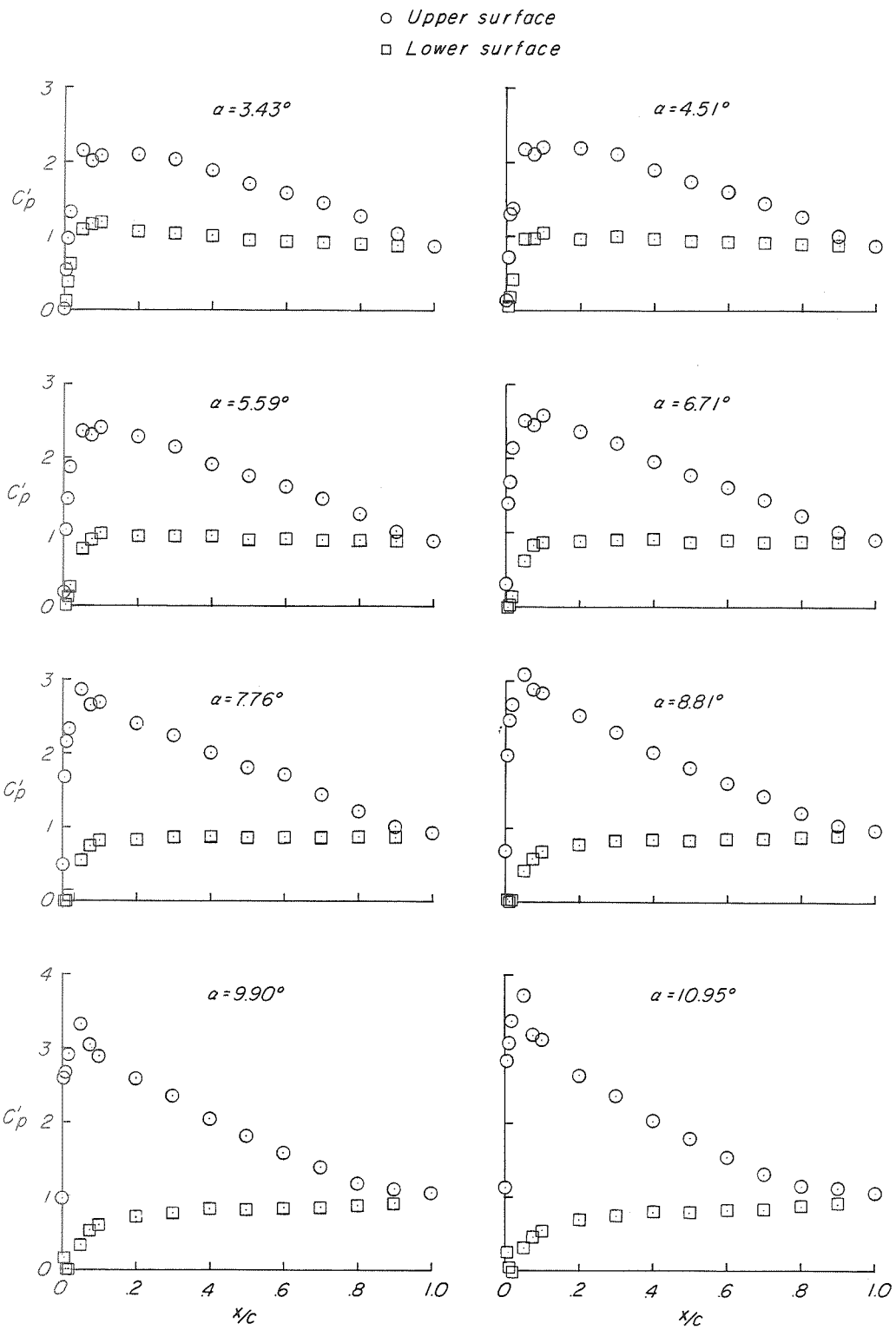
(d) $\alpha = 17.58^{\circ}$ to 20.59°.

Figure 7.- Concluded.



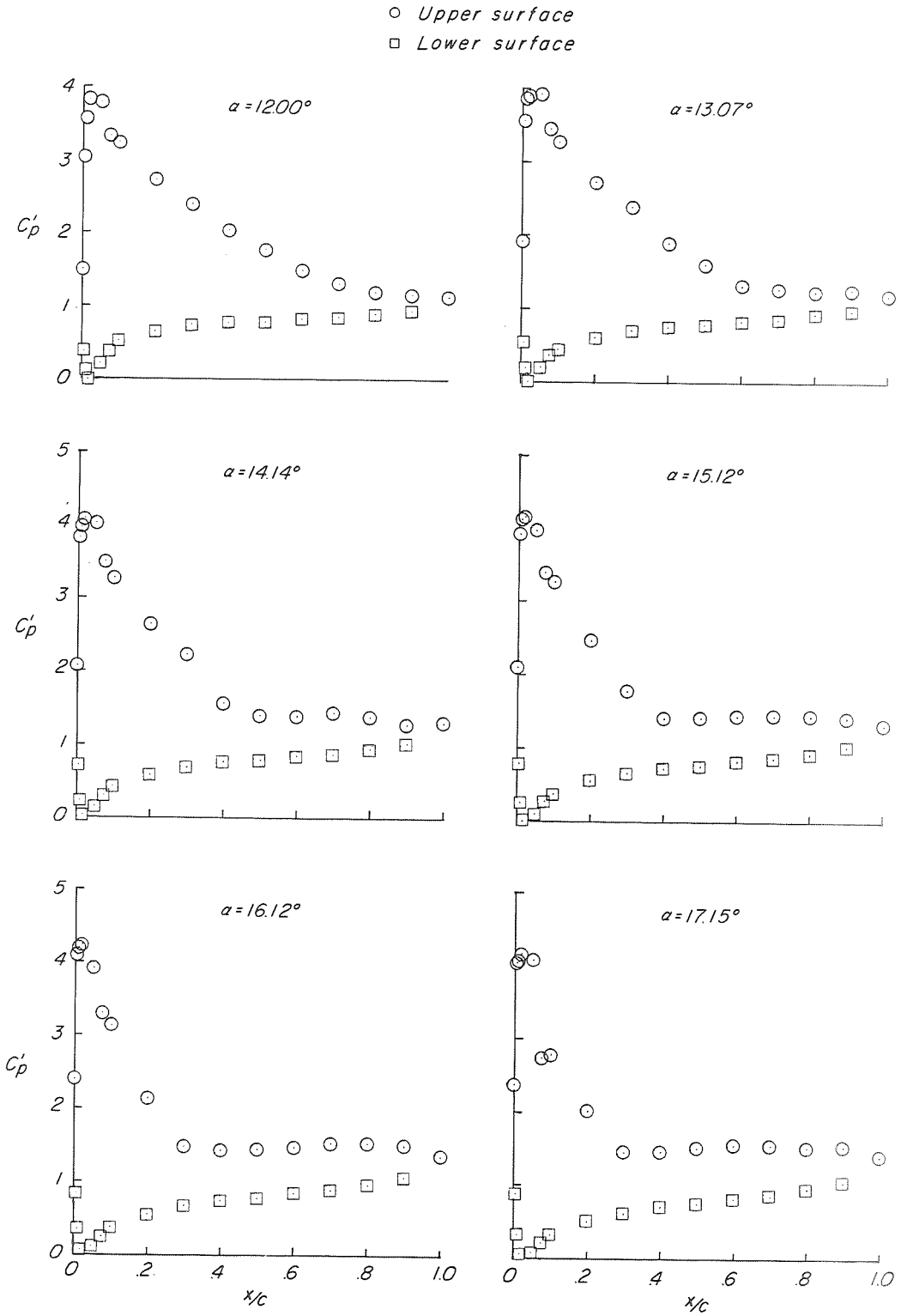
(a) $\alpha = -5.10^{\circ}$ to 2.34°.

Figure 8.- Pressure distributions on wing of model at free-stream dynamic pressure of approximately 2840 N/m^2 (59.3 lbf/ft^2). $\delta_f = 0^{\circ}$.



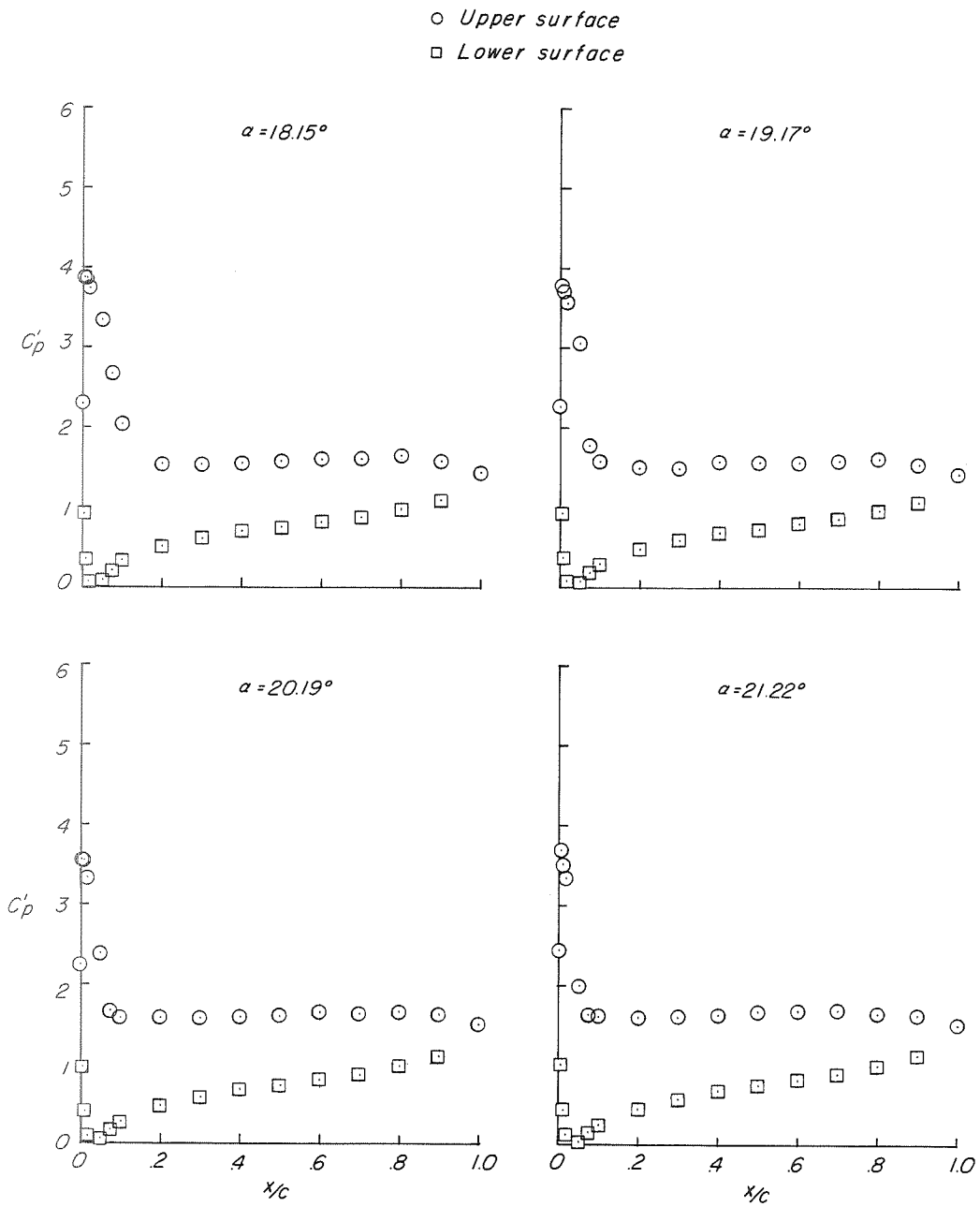
(b) $\alpha = 3.43^{\circ}$ to 10.95°.

Figure 8.- Continued.



(c) $\alpha = 12.00^{\circ}$ to 17.15°.

Figure 8.- Continued.



(d) $\alpha = 18.15^{\circ}$ to 21.22°.

Figure 8.- Concluded.

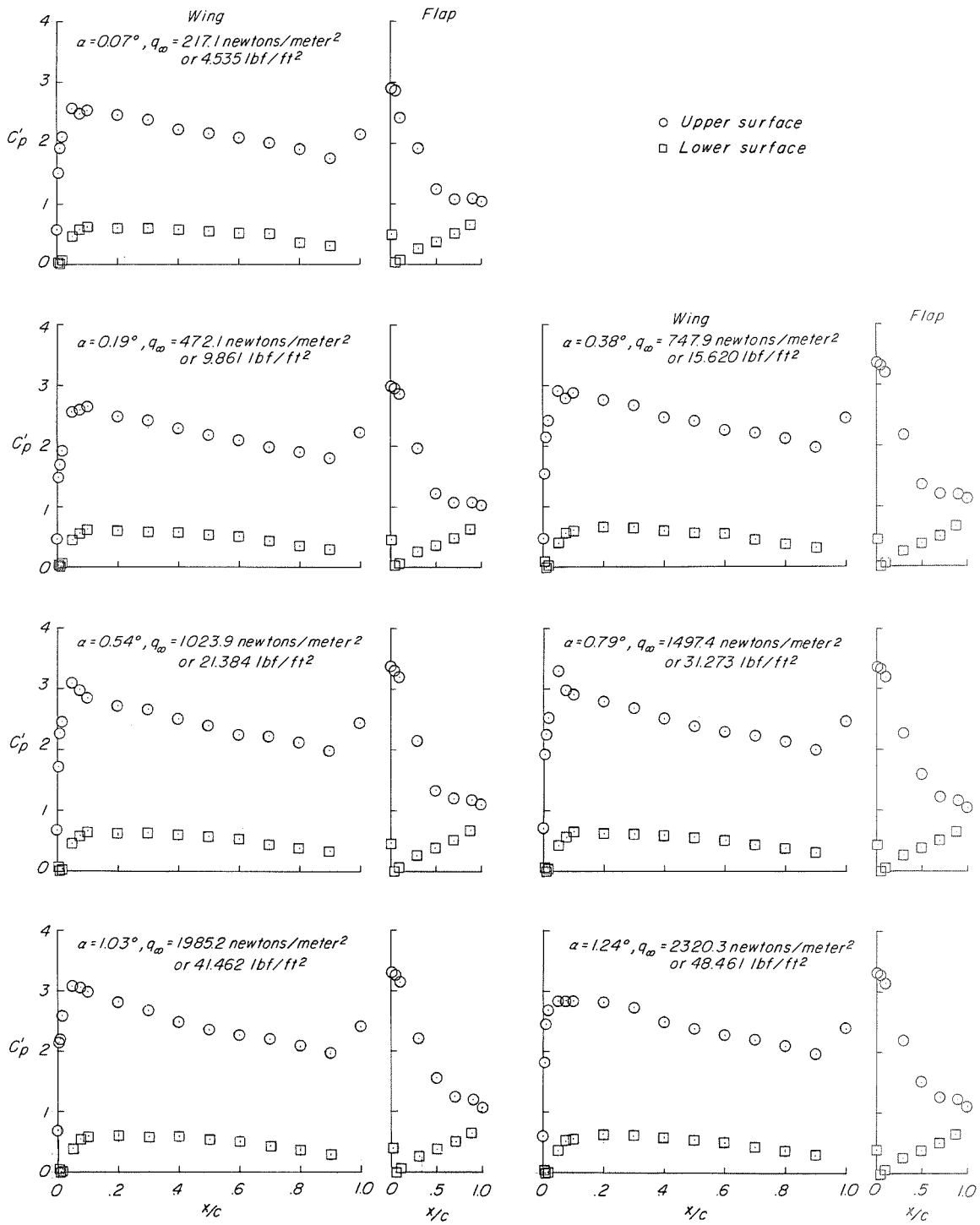
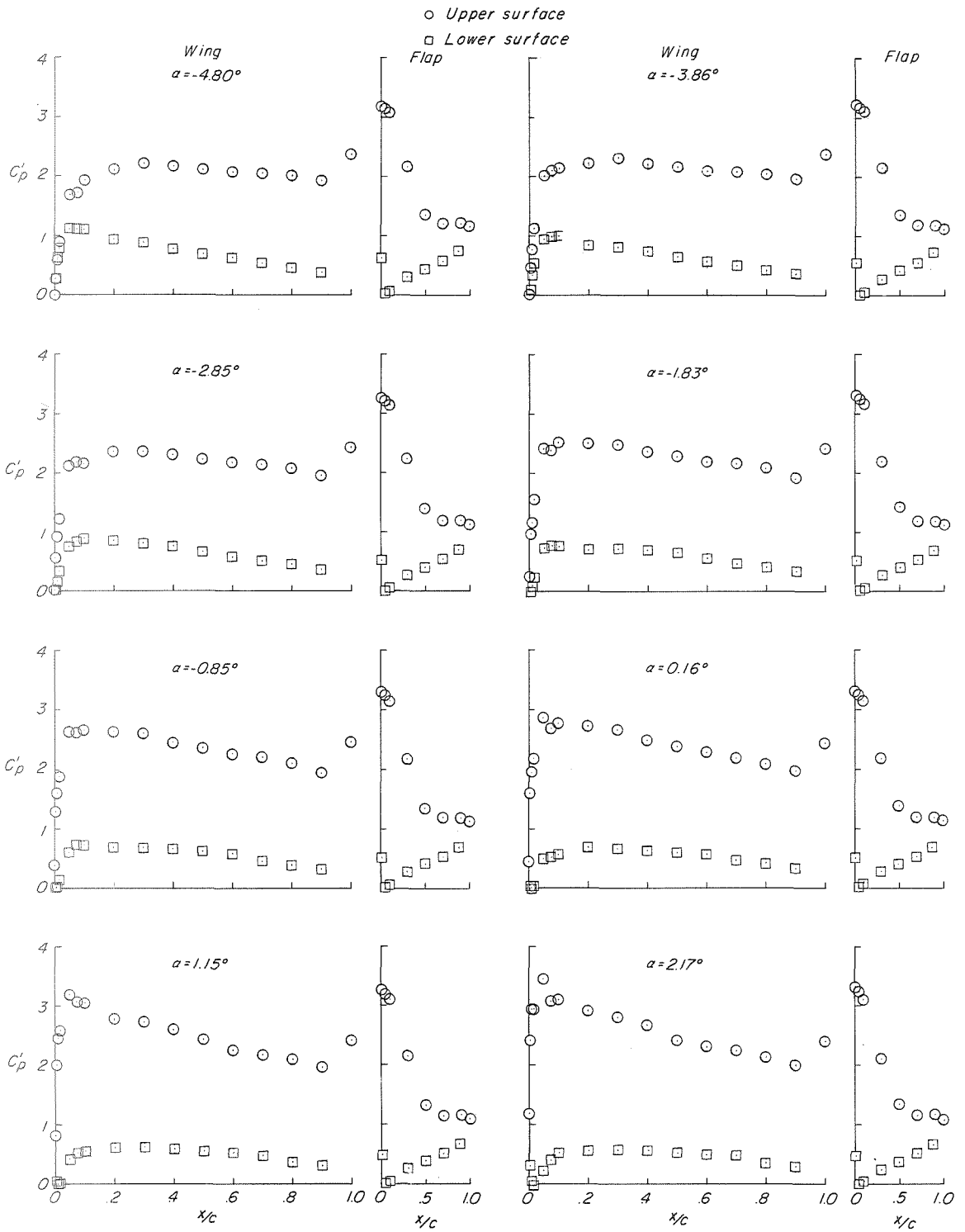
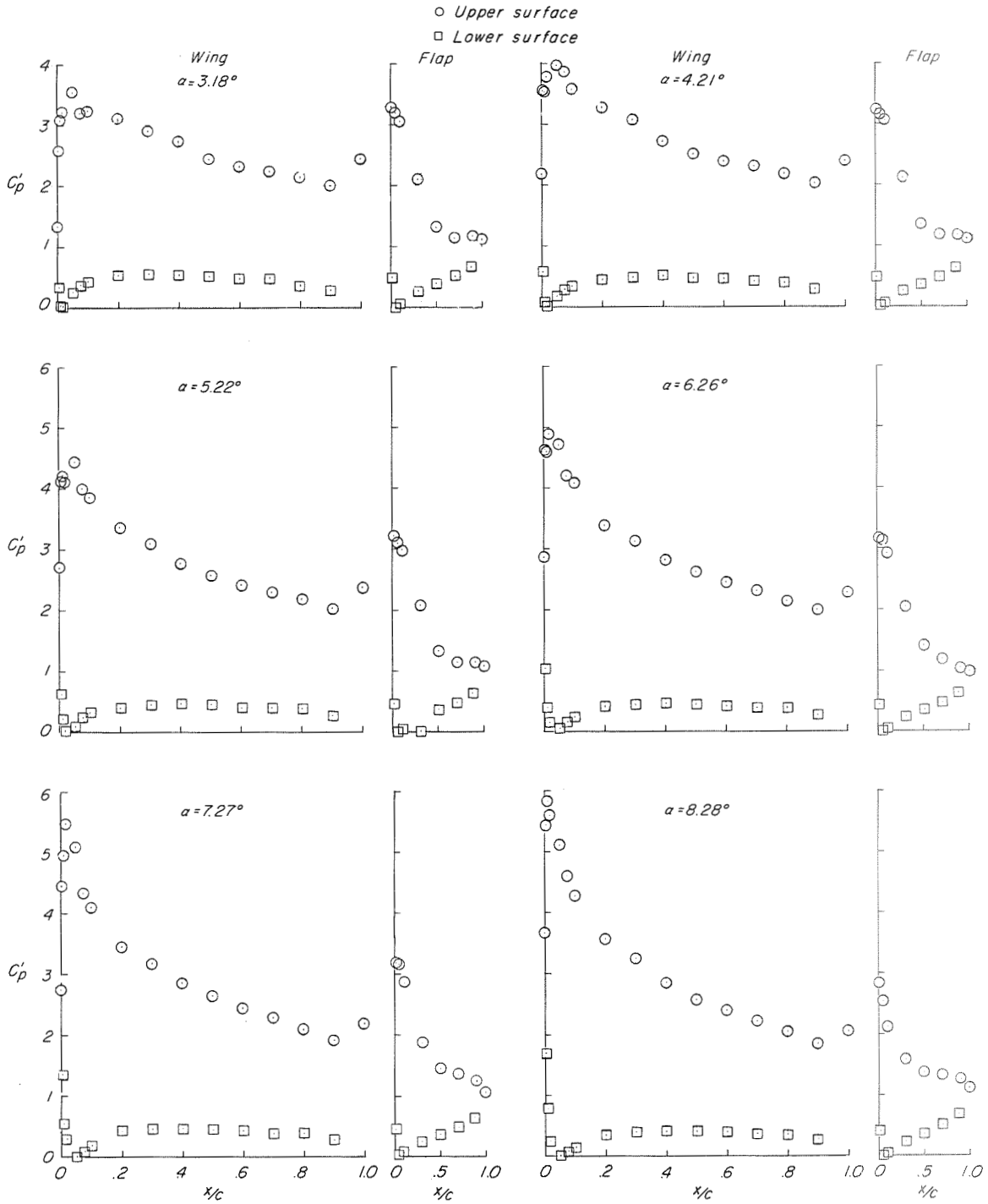


Figure 9.- Pressure distributions on wing and flap of model at several free-stream dynamic pressures. $\delta_f = 40^o$.



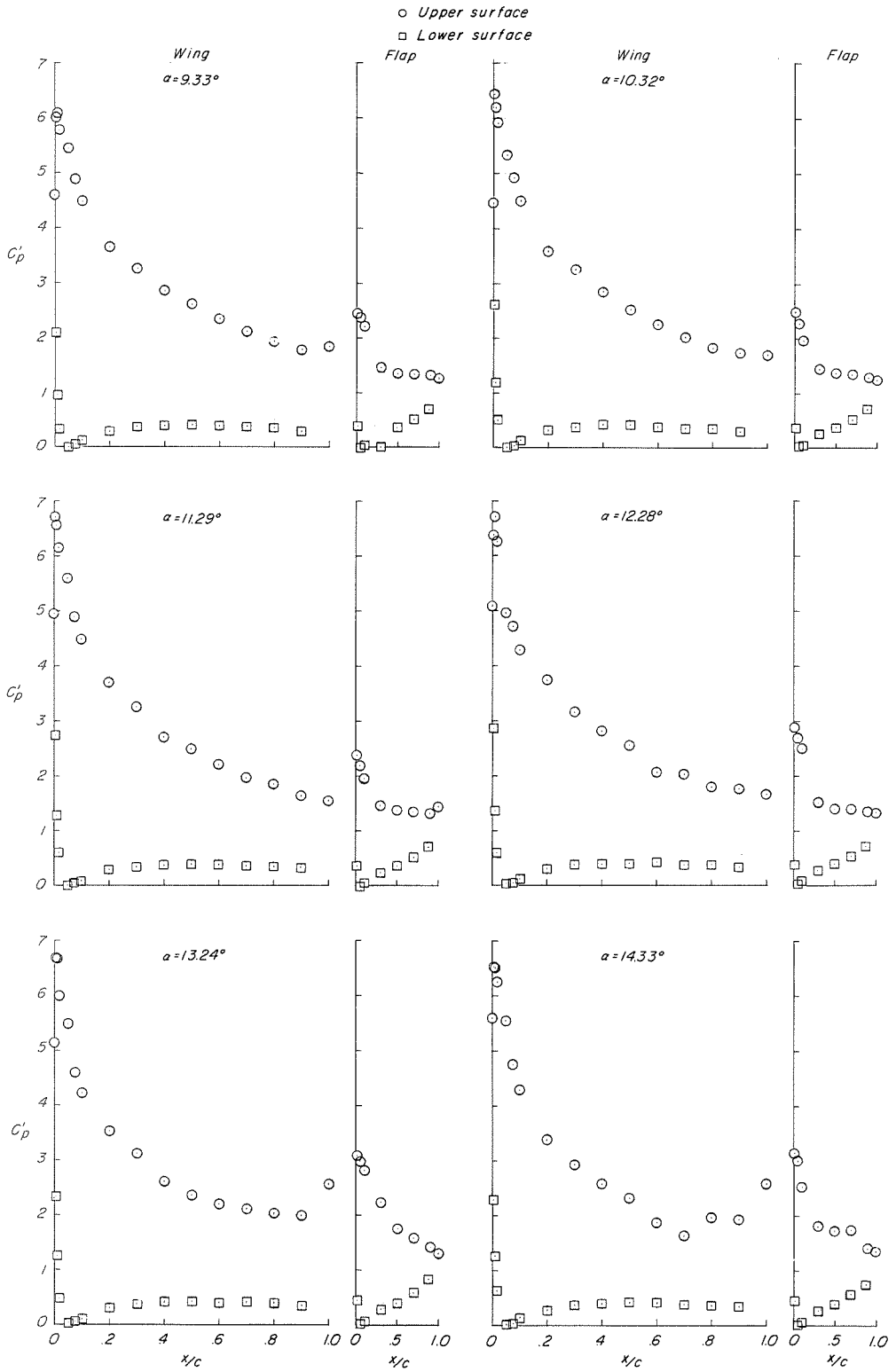
(a) $\alpha = -4.80^{\circ}$ to 2.17°.

Figure 10.- Pressure distributions on wing and flap of model at free-stream dynamic pressure of approximately 420 N/m^2 (8.8 lbf/ft^2). $\delta_f = 40^{\circ}$.



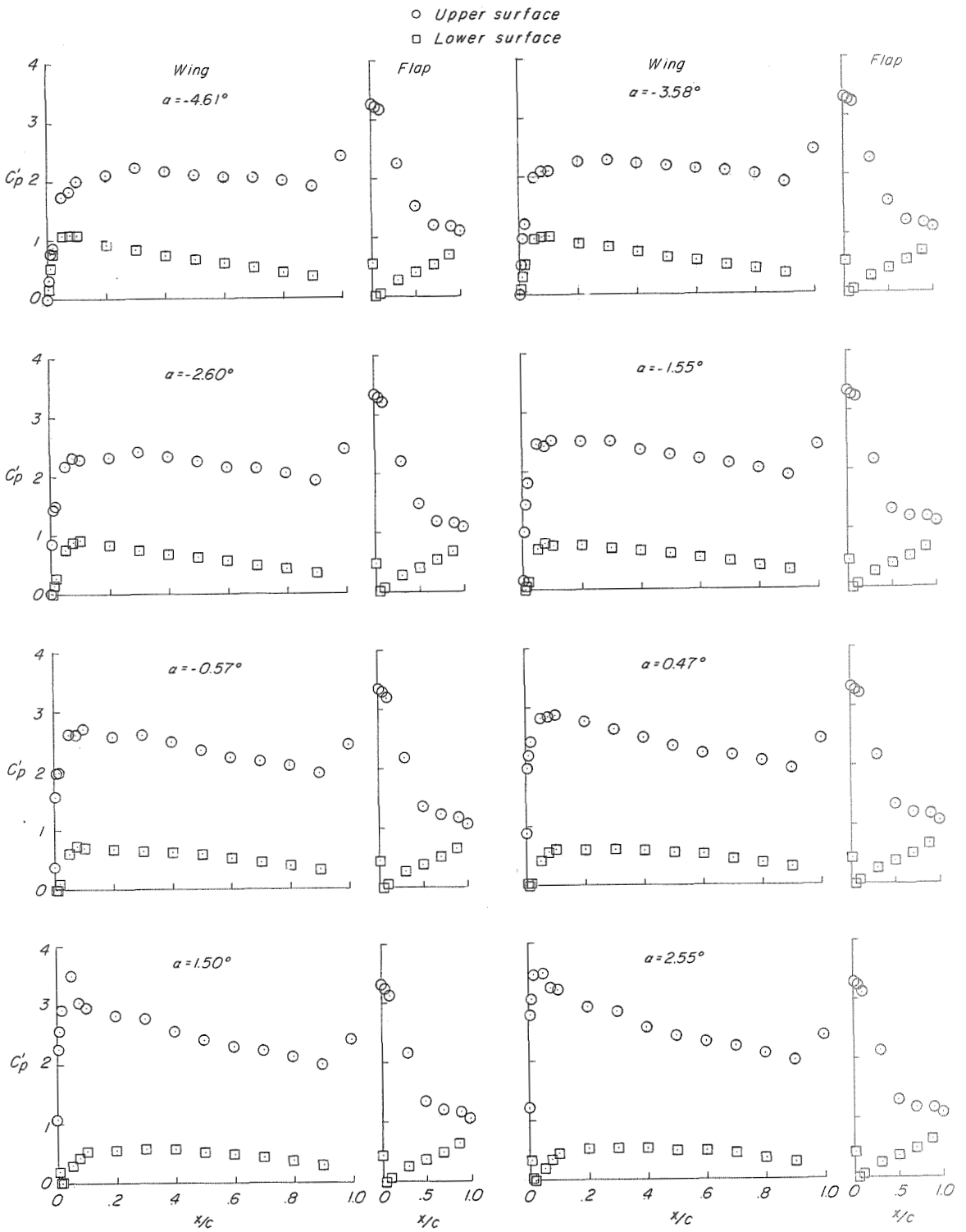
(b) $\alpha = 3.18^{\circ}$ to 8.28°.

Figure 10.- Continued.



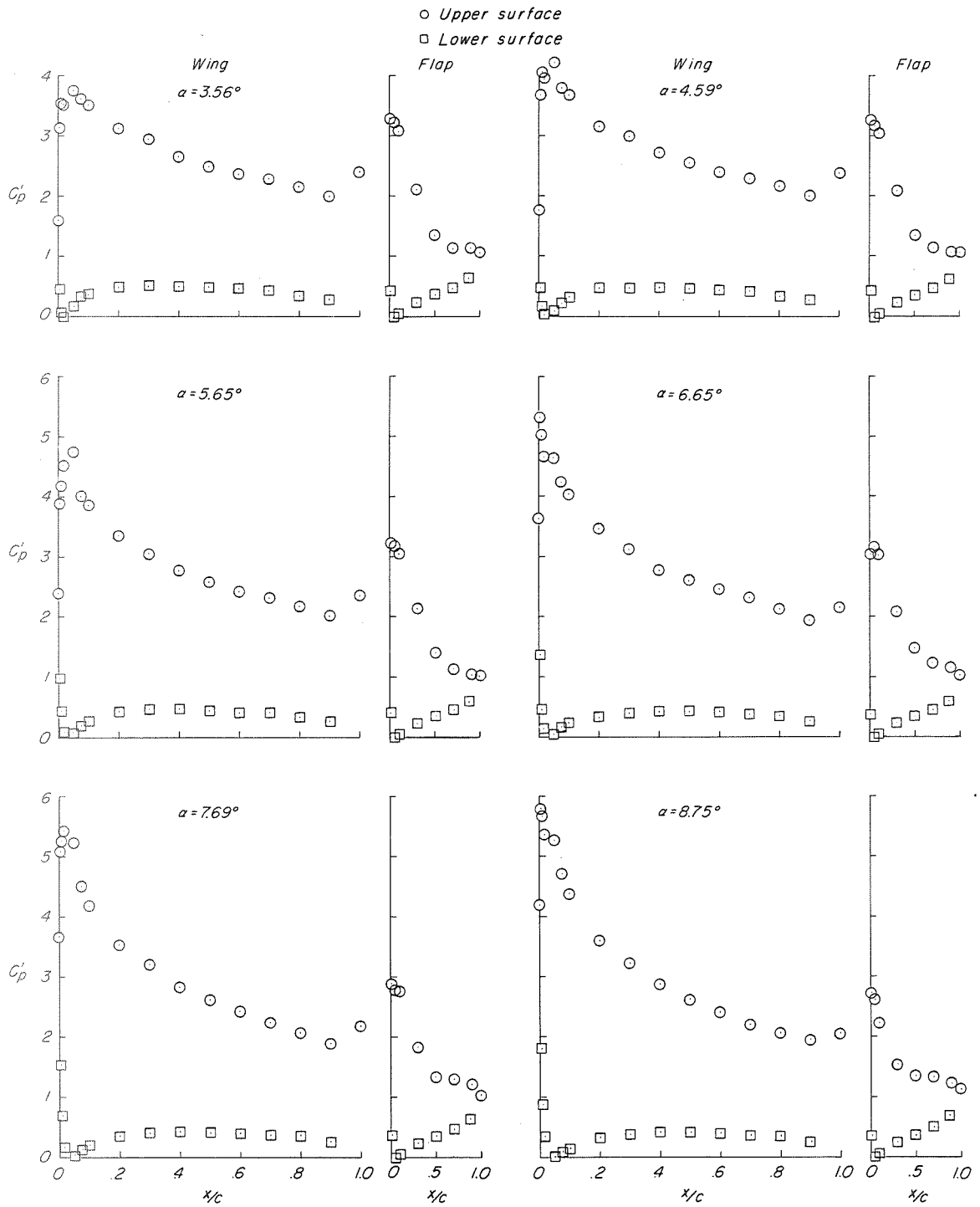
(c) $\alpha = 9.33^{\circ}$ to 14.33°.

Figure 10.- Concluded.



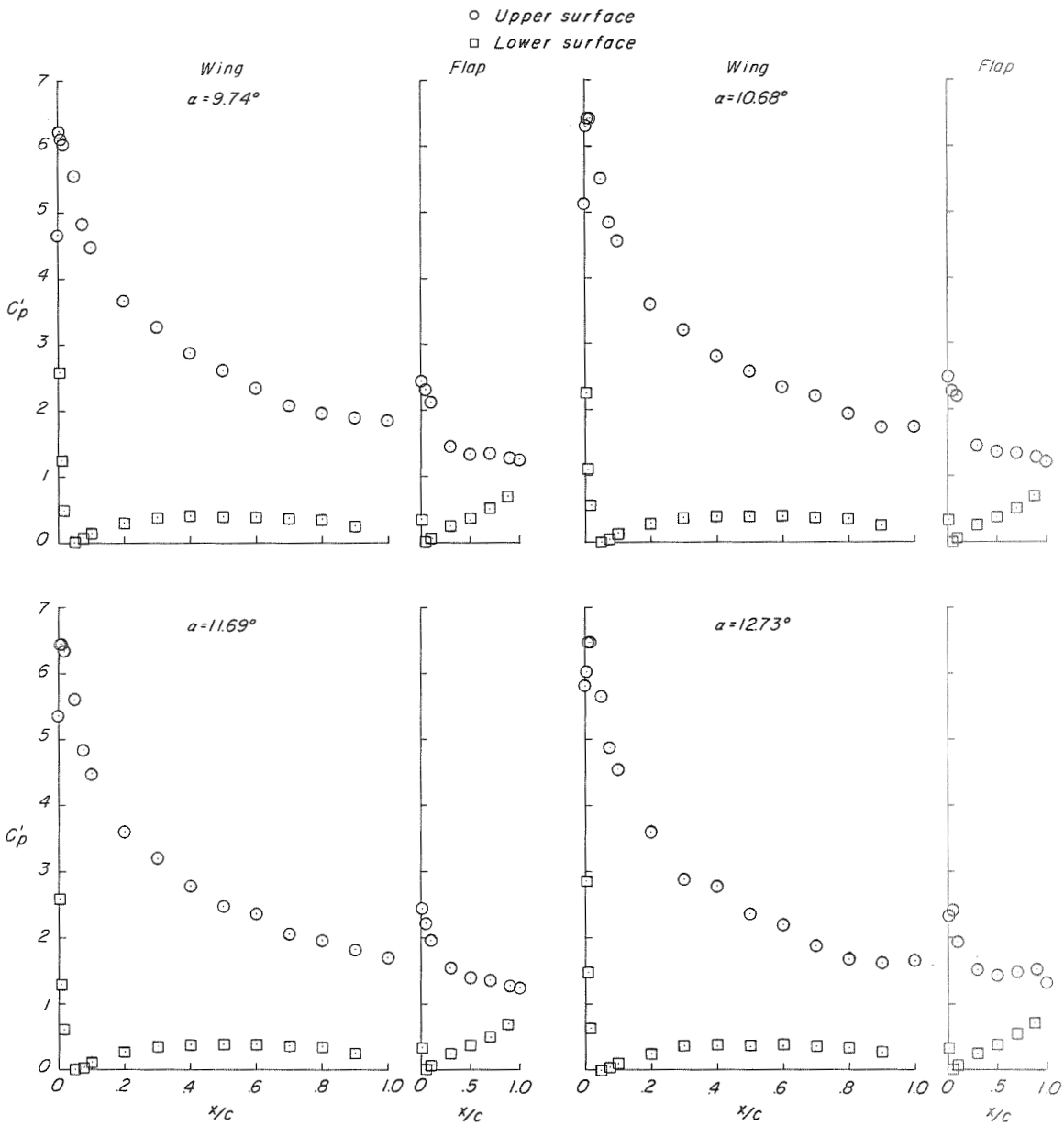
(a) $\alpha = -4.61^{\circ}$ to 2.55°.

Figure 11.- Pressure distributions on wing and flap of model at free-stream dynamic pressure of approximately 1020 N/m^2 (21.3 lbf/ft^2). $\delta_f = 40^{\circ}$.



(b) $\alpha = 3.56^{\circ}$ to 8.75°.

Figure 11.- Continued.



(c) $\alpha = 9.74^\circ$ to 12.73°.

Figure 11.- Concluded.

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