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PLOTIT-METHOD OF INTERACTIVELY PLOTTING  
INPUT DATA FOR THE VORLAX COMPUTER PROGRAM

(NASA C -158896) PLOTIT-METHOD OF  
INTERACTIVELY PLOTTING INPUT DATA FOR THE  
VORLAX COMPUTER PROGRAM (Vought Corp.,  
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**Langley Research Center**  
Hampton, Virginia 23665



## SUMMARY

A method of plotting the geometric input to the VORLAX computer program by means of an interactive remote computer terminal is described. The software consists of a procedure file and two programs and was developed for use with the Langley Research Center computer system. The programs and procedure file are described and a sample execution is presented.

## INTRODUCTION

The VORLAX computer program uses a sparse set of geometric input data to describe the aircraft configuration being analyzed. The geometry of the configuration can at times become very complex, and it is necessary to plot the configuration resulting from the data in order to ascertain its accuracy. The procedure file and two computer programs described herein provide a method for plotting this data at an interactive graphics terminal. A sample execution of the procedure is presented.

Use of this method allows the configuration to be plotted with any combination of roll, pitch, or yaw angles. Three independent forms of data display are available, and these may be specified in any combination. These are: (1) configurations with or without camber, (2) configurations showing only major panels or only minor panels, and (3) configurations with or without control points plotted. Any section of the plot may be enlarged for examination in greater detail.

The procedure file and computer programs to plot the VORLAX input data have been written to be used in the Langley Research Center computer system which provides a Network Operating System (NOS) and a Tektronix Plot 10 package. Langley Research Center users will find the procedure file and computer programs in mass storage as public files in the catalog of user number 214737C.

## DISCUSSION

### DESCRIPTION OF PROCEDURE FILES AND PROGRAMS

A method has been developed for plotting the VORLAX input data which consists of a procedure file and two computer programs. The procedure file PLOTIT and the computer programs READS and PLOTS are described in the following. A sample execution is provided which includes illustrations of the displays.

#### Procedure File PLOTIT

Procedure file PLOTIT (Appendix A) is used to simplify the plotting procedure. PLOTIT first gets the desired data file and renames it TAPE1. It then gets the binary form of program READS, which reads the input data and prepares files suitable for the subsequent operations. PLOTIT then gets and executes the binary form of PLOTS, which performs the aircraft geometry plotting.

#### Program READS

Program READS (Appendix B) reads the VORLAX input data from a disc file named TAPE1 and determines the necessary scaling factor in order for the aircraft drawing to fit on the screen.

The data for each of the panels are then read, and the coordinates of a set of points that describe the panels are calculated. These coordinates are stored on three disc files named TAPE3, TAPE4, and TAPE5. These data files will be read by program PLOTS.

Program READS provides the user with two methods of representing the aircraft. The first method displays the aircraft with camber as shown in figure 1(a). The second method shows the aircraft without camber. Figure 1(b) is the aircraft of figure 1(a) without camber.

#### Program PLOTS

Program PLOTS (Appendix C) reads the data produced by program READS. This program offers the user several variations in the display which are described

in the following sections.

Rotation of the Aircraft. - The initial position of the aircraft is a side view with the nose to the left. From this position, it is possible to rotate the aircraft first in roll, then in pitch, and finally in yaw. Positive angles are defined as follows: roll, right wing down; pitch, nose up; and yaw, nose right.

Additional Independent Variations. - PLOTS offers two additional independent variations in the display. These variations are: (1) with or without subpaneling and (2) with or without control points. Examples of these variations are shown in figures 1(c) through 1(f). Figure 1(c) shows the aircraft without subpaneling or control points; figure 1(d) has subpaneling added; in figure 1(e) control points only have been added; and in figure 1(f) control points and subpaneling have been added. All of the plots are constructed using orthographic projection.

Recovery from Input Errors. - If an error has been made in specifying the input parameters, the execution of PLOTS can be stopped. There are three ways to stop PLOTS while it is executing. These are: (1) enter a value greater than 360 for roll angle, (2) answer any of the questions with STOP, or (3) stop the program while it is plotting by interrupting it with the break key and then entering an S.

Program PLOTS can be restarted at the beginning by sending the command PLOTB to the computer. PLOTB will also restart the program if it is stopped for any other reason.

#### SAMPLE EXECUTION

Figure 2 shows a sample execution of PLOTIT at a remote terminal. The first command gets the procedure file PLOTIT.

```
GET,PLOTIT/UN=214737C
```

The second command initiates the execution of the procedure file.

CALL,PLOTIT(T=VORLAXX)

VORLAXX is the name of the file on which the VORLAX data deck has been stored for this example. The file name in the calling statement can be any name which corresponds to a file on which VORLAX data is stored.

The first question asked by the computer deals with camber in the panels.

DO YOU WANT CAMBER IN THE PANELS ?  
TRUE OR FALSE

If camber is desired in the panels, type in TRUE, otherwise type in FALSE and the camber will be set equal to zero.

The next three questions asked by the computer are concerned with the desired roll, pitch, and yaw angles of the configuration, and are as follows.

INPUT THE ROLL ANGLE FOR THE AIRCRAFT  
(DEG), > 360 TO STOP.  
PITCH ANGLE  
YAW ANGLE

The angles desired in degrees, are typed in after the questions. If termination of the program is desired, a value greater than 360 may be typed in for the roll angle.

The next two questions are concerned with the desirability of displaying paneling and control points. The two questions are:

SUBPANELING ? TRUE OR FALSE  
CONTROL POINTS ? TRUE OR FALSE

If these questions are answered TRUE, the subpaneling and control points are incorporated into the plots. If they are answered FALSE, then these quantities are deleted. The plot resulting from the input in figure 2 is presented in figure 3.

If a certain section of the plot needs to be enlarged in order to examine the plotted data more closely, this may be accomplished at the terminal. When the computer has finished plotting, it will print the following statement:

FOR ENLARGEMENT INPUT YES

At this point a hard copy can be made if desired. If any reply except YES is typed in, the computer will ask for a new set of angles. If YES is typed in, the graphics cursor (cross hairs) will appear. The cursor should then be located at the lower left corner [fig. 4(a)] of a rectangular region to be enlarged. A non-control keyboard character should be pressed. This will cause the cursor to disappear. The carriage return is then pressed. This sequence sends the coordinates of the first corner to the computer. The graphics cursor will reappear and should be relocated to the upper right corner of the desired rectangular region [fig. 4(b)]. A non-control keyboard character and the carriage return are then pressed as for the first corner. An enlargement of the region defined by these positions of the graphics cursor is shown in figure 4(c).

When the plot is finished, the computer will again print

·FOR ENLARGEMENT INPUT YES

This allows a further enlargement of a section of the plot if desired.

#### CONCLUDING REMARKS

A plotting routine, PLOTIT, has been developed for plotting the input data for the VORLAX computer program. This program allows the user to plot geometric input data interactively at a remote graphics terminal and thereby ascertain very rapidly whether or not the data is correct.

The routine consists of two programs and a procedure file. These have been designed for use on the Control Data Corporation computer system with a Network Operating System (NOS) and a Tektronix Plot 10 graphics package at the NASA Langley Research Center.

REFERENCES

1. Miranda, Luis R.; Elliott, Robert D.; and Baker, William M.: A Generalized Vortex Lattice Method for Subsonic and Supersonic Flow Applications. NASA CR-2865, December 1977.

## APPENDIX A

### PROCEDURE FILE PLOTIT

This procedure file gets the binary form of READS(READB) and executes it, then gets the binary form of PLOTS (PLOTB) and executes it.

```
PLOTIT.  
RETURN,TAPE3,TAPE4,TAPE5.  
GET,TAPE1=T.  
GET,READB/UN=214737C.  
READB.  
RETURN,READB.  
GET,PLOTB/UN=214737C.  
PLOTB.  
EXIT.
```

## APPENDIX B

## SOURCE LISTING OF PROGRAM READS

```

C      PROGRAM READS(INPUT,OUTPUT,TAPE1,TAPE4,TAPE2=OUTPUT,TAPE3,TAPE5)
C      THIS PROGRAM READS THE INPUT DATA FOR THE VORLAX PROGRAM
C      SO IT CAN BE PLOTTED.
      DIMENSION TITLE(8)
      COMMON/BLOCK/XOFFSET
      LOGICAL CAMBER
      REAL LENGTH
      READ(1,100)TITLE
      READ(1,103)LAX,LAY
      READ(1,101)
      READ(1,102)NPAN,WSPAND
      CALL SIZES(NPAN,LENGTH,XOFFSET,WSPAN)
      IF(WSPAND.GT.WSPAN)WSPAN=WSPAND
      TESTRTD=LENGTH/WSPAN
      IF(LENGTH.GT.WSPAN)WSPAN=LENGTH
      REALY IS THE WIDTH OF THE PLOTTING SURFACE.
      WRITE(3)NPAN,TITLE,TESTRTD
      REALY=10.0
      REALY=REALY-.20
      SCALE=REALY/WSPAN
      XOFFSET=-(XOFFSET+WSPAN*.5)
      WRITE(2,104)
      READ 105,CAMBER
      DO 200 I=1,NPAN
      200 CALL PANLRED(SCALE,LAX,LAY,CAMBER)
      100 FORMAT(8A10)
      101 FORMAT(//)
      102 FORMAT(I2,48X,F10.0)
      103 FORMAT(I1X,I1,9X,I1)
      104 FORMAT("DO YOU WANT CAMBER IN THE PANELS ? ",/, "TRUE OR FALSE")
      105 FORMAT(L7)
      STOP
      END

```

1 READS  
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APPENDIX B. - Continued

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 71 READS

```

SUBROUTINE PANLRED(SCALE,LAX,LAY,CAMBER)
COMMON/BLOCK/XOFFSET
COMMON/BLOKO/NAP,XAF(50),ZC(2,50),CORD1,CORD2,CD1SIN,CD2SIN
COMMON/CANDS/SIN1,SIN2,COS1,COS2,DELTA
COMMON/ROJ1/ROJ,CROSSIZ
LOGICAL CAMBER
INTEGER PRD,RNCV

C
C THIS SUBROUTINE READS THE VORLAX DATA FOR A PANEL, EACH TIME IT
C IS CALLED. THE COORDINATES OF A SET OF POINTS THAT DESCRIBE
C THE PANEL ARE CALCULATED AND STORED ON DISC.
C
DIMENSION X(2),Y(2),Z(2),CORD(2),RLE(2)
COMMON/PHIRO/ PHI(100),RO(100),SINE(100),COSINE(100)
COMMON VORS(3,500)
COMMON/PI/PIE
COMMON/TWIST/AINC1,DAINC
REAL K
DATA PIE/3.14159/,CROSSIZ/.02/

C
C THIS SECTION READS THE VORLAX DATA CARDS FOR A PANEL.
C
DO 200 I=1,2
200 READ(1,100)X(I),Y(I),Z(I),CORD(I)
DO 250 I=1,2
X(I)=(X(I)+XOFFSET)*SCALE
Y(I)=Y(I)*SCALE
Z(I)=Z(I)*SCALE
250 CORD(I)=CORD(I)*SCALE
READ(1,101)TVOR,TNCV,PDL
RNCV=INT(TNCV)
NVOR=INT(TVOR)
NVOR1=NVOR+1
IF(PDL.LE.360.)GO TO 1
READ(1,102)(PHI(N),RO(N),N=1,NVOR1)
DO 251 N=1,NVOR1
PHI(N)=PHI(N)*3.14159/180.
251 RO(N)=RO(N)*SCALE
    
```

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APPENDIX B. - Continued

111 READS  
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 147 READS

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Y1=Y(1)
CORD2=CORD(2)
SIN1=SIN(AINC1)
SIN2=SIN(AINC2)
COS1=COS(AINC1)
COS2=COS(AINC2)
CD1SIN=CORD1*SIN1
CD1COS=CORD1*COS1
CD2SIN=CORD2*SIN2
CD2COS=CORD2*COS2
Z1=Z(1)
Z2=Z(2)
DELTAY=(Y(2)-Y1)
DELTAX=(X2-X1)
DELTAZ=(Z2-Z1)
DELTACO=(CD2COS-CD1COS)
YRAT=SQRT(DELTAX*DELTAX+DELTAY*DELTAY)

C
ISETS=ISETSF(NVRNPRD)
CALL SAVE(IFUG,3,IDUM,1)
WRITE(3)NVOR,RNCV,PDL,IQUANT,NVRNPRD,ISETS
CALL INTERP(0.0,ZZ,ZZ,X00,XERP)
DO 310 I=1,NVOR1
IF(LAY.EQ.1)RATIO=(I-1.)/NVOR
IF(LAY.EQ.0)RATIO=.5*(1.-COS(PIE*(I-1.)/NVOR))
VORS(1,I)=X1+X00+RATIO*(DELTAX+XERP)
VORS(2,I)=Y1+DELTAY*RATIO
VORS(3,I)=ZZ+RATIO*(ZZ+DELTAZ)+Z1
310

C
DO 501 I=NVOR1,NVRN,NVOR1
IF(LAY.EQ.1)PCORD=(I/NVOR1*4.0-3.0)/N4
IF(LAY.EQ.0)PCORD=.5*(1.-COS(2.*I/NVOR1-1.))*PIE/N2)
CALL INTERP(PCORD,ZZO,ZZ,X00,XERP)
DO 501 J=1,NVOR1
I2=I+J+IFUG
IF(LAY.EQ.1)KATIO=(J-1.)/NVOR
IF(LAY.EQ.0)RATIO=.5*(1.-COS(PIE*(J-1.)/NVOR))

```

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APPENDIX B. - Continued

```

1300 IF(I2.EQ.500)CALL SAVE(IFUG,5, IDUM, 2)
      CALL SAVE(IFUG,5, NVRNPRD, 3)
C
C   THIS SECTION CALCULATES THE LOCATION OF A SET OF POINTS THAT
C   DESCRIBE A CURVED MAJOR PANEL, WITHOUT SUBPANELING.
      NVRN=(NAP)*(NVOR+1)
      NVRNPRD=(NAP)*(NVOR+1)
      ISETS=ISETSF(NVRNPRD)
      WRITE(3)NVOR , NAP , PDL, IQUANT, NVRNPRD, ISETS
      CALL SAVE(IFUG, 3, NVRNPRD, 1)
      IF(NAP.LE.2)GO TO 556
      DO 555 I=1, NVRN, NVOR1
      II=(I)/NVOR1+1
      XTEMP=X1+CORD1*XAF(II)
      SKAL=ZC(2,II)
      DO 555 J=1, NVOR1
      R=RO(J)*SKAL
      FI=PHI(J)
      I2=I+J-1+IFUG
      VORS(1, I2)=XTEMP
      VORS(2, I2)=R*COS(FI)+YNOT
      VORS(3, I2)=R*SIN(FI)+ZNOT+CORD1*.01*ZC(1, II)
555  IF(I2.EQ.500)CALL SAVE(IFUG, 3, IDUM, 2)
556  IF(NAP.GT.2)GO TO 5555
      NAP=2
      NVRNPRD=NAP*NVOPI
      DO 5554 J=1, NVOR1
      J2=J+NVOR1
      VOPS(1, J)=X1
      VORS(1, J2)=X1+CORD1
      VORS(2, J)=VORS(2, J2)-YNOT+RO(J)*COS(PHI(J))
      VORS(3, J)=VORS(3, J2)-ZNOT+RO(J)*SIN(PHI(J))
5554 CALL SAVE(IFUG, 3, NVRNPRD, 3)
5555
C   THIS SECTION CALCULATES THE LOCATION OF THE CONTROL POINTS FOR
C   A CURVED MAJOR PANEL. IT THEN PLACES AN "X" ON EACH POINT.
      POJ=PO(1)
      DO 2010 N=1, NVOR

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APPENDIX B. - Continued

300 READS  
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 337 READS

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AZO=RO(N)*SIN(PHI(N))
AZ1=RO(N+1)*SIN(PHI(N+1))
AYO=RO(N)*COS(PHI(N))
AY1=RO(N+1)*COS(PHI(N+1))
DAZ=AZO-AZ1
DAY=AYO-AY1
H=SQRT(DAZ*DAZ+DAY*DAY)
SINE(N)=DAZ/H
COSINE(N)=DAY/H
R1=RO(N)
R2=RO(N+1)
F1=PHI(N)
F2=PHI(N+1)
RO(N)=(R1*SIN(F1)+R2*SIN(F2))*0.5
PHI(N)=(R1*COS(F1)+R2*COS(F2))*0.5
2010 PRD=NVDOR*RNCV
      NNO=(RNCV-1)*NVOR+1
      NNN=PRD*4
      ISETS=ISETSF(NNN)
      WRITE(4)NVOR,RNCV,PRD,NNN,NNN,NNO,ISETS
      CALL SAVE(IFUG,4,IDUM,1)
      DO 2001 I=1,NNO,NVOR
      K=(I-1)/NVOR+1
      IF(LAX.EQ.0)PCORD=.5*(1.-COS(K*PIE/RNCV))
      IF(LAX.EQ.1)PCORD=(4.*K-1.)/N4
      CALL INTERP2(PCORD,TEMX,SKAL,DSKL,CROSSX,CAMB,DZ,6HPOINTS)
      TEMX=TEMX*X1
      DO 2001 J=1,NVOR
      J2=(I+J-2)*4+IFUG
      J21=J2+1
      J22=J2+2
      J23=J2+3
      J24=J2+4
      VS=SINE(J)*CRDSSIZ
      HS=COSINE(J)*CROSSIZ
      TEMY=PHI(J)*SKAL+YNOT
      TY2=DSKL*PHI(J)
      TEMZ=RO(J)*SKAL+ZNOT+CAMB
    
```

APPENDIX B. - Continued

338 READS  
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 355 READS

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TZ2=RO(J)*DSKL+DZ
VORS(1,J21)=TEMX-CROSSX
VORS(1,J22)=TEMX+CROSSX
VORS(1,J23)=TEMX
VORS(1,J24)=TEMX
VORS(2,J21)=TEMY-TY2
VORS(2,J22)=TEMY+TY2
VORS(2,J23)=TEMY-HS
VORS(2,J24)=TEMY+HS
VORS(3,J21)=TEI1-TZ2
VORS(3,J22)=TEI1+TZ2
VORS(3,J23)=TEI1-VS
VORS(3,J24)=TEI1+VS
IF(J24.EQ.500)CALL SAVE(IFUG,4,IOUM,2)
2001 CONTINUE
CALL SAVE(IFUG,4,NNN,3)
RETURN
END
    
```

356 READS  
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 373 READS

```

SUBROUTINE INTERP(P,ZA,Z,XA,X)
C THIS SUBROUTINE DOES THE INTERPOLATION FOR CAMBER OF A FLAT
C MAJOR PANEL.
COMMON/BLOKO/NAP,XAF(50),ZC(2,50),CORD1,COR02,COSIN,COS2SIN
COMMON/CANDS/SIN1,SIN2,COS1,COS2,DELTA
COMMON/TWIST/AINC1,DAINC
COMMON/SLOPE/ANGL,DANG
IF(NAP.LE.2.OR.DELTA.EQ.0.0)GO TO 3
DO 1 I=2,NAP
1 IF(P.LE.XAF(I))GO TO 2
2 DAF=XAF(I)-XAF(I-1)
DZC1=ZC(1,I)-ZC(1,I-1)
DZC2=ZC(2,I)-ZC(2,I-1)
A1=ATAN(DZC1/DAF*.01)
A2=ATAN(DZC2/DAF*.01)
DAA=A2-A1
PC=(P-XAF(I-1))/DAF
CAM91=(DZC1*PC+ZC(1,I-1))*CORD1*.01
    
```

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APPENDIX B. - Continued

READS 410  
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 READS 441

```
PC2=(P+P2-XAF(I-1))/DAF
S2=ZC(2,I-1)+DZC2*PC2
Z2=ZC(1,I-1)+DZC1*PC2
Z2=Z2*CORD1*.01
DZ=Z2-Z
DS=S2-S
RETURN
3 S=S2-1.0
Z=DS=DZ=0.0
CROS=CROSSIZ
RETURN
END
```

```
SUBROUTINE ANGLE(R,S,C)
THIS SUBROUTINE IS USED FOR PLACING X'S ON CONTROL POINTS,
ON FLAT MAJOR PANELS.
COMMON/SLOPE/ANGL,DANG
ANG=ANGL+R*DANG
S=SIN(ANG)
C=COS(ANG)
RETURN
END
```

```
SUBROUTINE ZEROZG1(POL)
THIS SUBROUTINE REMOVES CAMBER FROM THE MAJOR PANELS.
COMMON/BLDKO/NAP,XAF(50),ZC(2,50),CORD1,CORD2,CD1SIN,CD2SIN
DO 1 I=1,2
IF(I.EQ.2,A.POL.GE.360.)RETURN
DO 1 J=1,NAP
1 ZC(I,J)=0.0
RETURN
END
```

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APPENDIX B. - Continued

442	READS		
443	READS		
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446	READS		
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469	READS		
470	READS		
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472	READS		
473	READS		
474	READS		
475	READS		

```

C      FUNCTION ISETSF(N)
C      THIS FUNCTION DETERMINES THE NUMBER OF DATA SETS NEEDED TO
C      TC DESCRIBE A MAJOR PANNEL.
      R=1.0/500.*N
      ISETSF=INT(R)
      IF(ISETSF.LT.R)ISETSF=ISETSF+1
      RETURN
      END
C

```

```

C      SUBROUTINE SAVE(FUG,FILE,PRJD,TEST)
C      THIS SUBROUTINE MAKES IT POSSIBLE TO REDUCE CORE REQUIREMENTS
C      BY REDUCING ARRAY SIZE.
      INTEGER FUG,FILE,PNTS,TEST,PROD
      COMMON VORS(3,500)
      IF(TEST.EQ.2)GO TO 2
      IF(TEST.EQ.3)GO TO 3
      FUG=0
      RETURN
2     FUG=FUG-500
      WRITE(FILE)((VORS(I,J),I=1,3),J=1,500)
      RETURN
3     PNTS=MOD(PROD,500)
      IF(PNTS.EQ.0)RETURN
      WRITE(FILE)((VORS(I,J),I=1,3),J=1,PNTS)
      RETURN
      END
C

```

```

C **  SUBROUTINE SIZES(NPAN,LENGTH,XOFFSET,WIDTH)
C **  DETERMINES THE LENGTH, WIDTH, STARTING POINT OF THE
      AIRCRAFT.
      REAL LENGTH
      NPAN2=NPAN-1
      CALL FIND(LENGTH,H,XOFFSET,WIDTH)
      IF(NPAN.EQ.1)GO TO 8

```

APPENDIX B. - Continued

476 READS  
 477 READS  
 478 READS  
 479 READS  
 480 READS  
 481 READS  
 482 READS  
 483 READS  
 484 READS  
 485 READS  
 486 READS  
 487 READS  
 488 READS

```

00 7 I=1,NPANZ
    CALL FIND(BIG,SMALL,WIDE)
    IF(SMALL.LT.XOFFSET)XOFFSET=SMALL
    IF(WIDE.GT.WIDTH)WIDTH=WIDE
    7 IF(BIG.GT.LENGTH)LENGTH=BIG
      WIDTH=WIDTH*2.0
      LENGTH=LENGTH-XOFFSET
    8 REWIND 1
      READ(1,5)
    5 FORMAT(////)
    RETURN
    END
    
```

C

489 READS  
 490 READS  
 491 READS  
 492 READS  
 493 READS  
 494 READS  
 495 READS  
 496 READS  
 497 READS  
 498 READS  
 499 READS  
 500 READS  
 501 READS  
 502 READS  
 503 READS  
 504 READS  
 505 READS  
 506 READS  
 507 READS  
 508 READS  
 509 READS  
 510 READS

```

SUBROUTINE FIND(BIG,SMALL,WIDE)
  DETERMINES THE STARTING POINT, END POINT, AND
  DISTANCE FROM THE AXIS FOR A PANEL.
  COMMON/BLOK0/NAP,XAF(50),ZC(2,50),CORD1,CORD2,CD2SIN,CD2SIN
  COMMON/PHIRD/ PHI(100),RD(100),SINE(100),COSINE(100)
  READ(1,100)X1,Y1,Z1,CORD1
  READ(1,100)X2,Y2,Z2,CORD2
  Y1=ABS(Y1)
  Y2=ABS(Y2)
  READ(1,101)TVOR,TNCV,PDL
  NVOR1=INT(TVOR)+1
  IF(PDL.LE.360.)GO TO 1
  READ(1,102)(PHI(N),RO(N),N=1,NVOR1)
  1 READ(1,103)AINC1,AINC2,ITS,NAP,IQUANT,ISYNT,NPP
  IF(ISYNT.NE.0)READ(1,104)
  IF(NAP.LE.2)GO TO 2
  READ(1,102)(XAF(I),I=1,NAP)
  IF(ITS.EQ.0.OR.PDL.GE.360.)GO TO 3
  READ(1,105)R
  3 READ(1,102)(ZC(1,I),I=1,NAP)
  IF(ITS.EQ.0.OR.PDL.GT.360.)GO TO 4
  READ(1,105)R
    
```

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APPENDIX B. - Concluded

READS	511
READS	512
READS	513
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READS	518
READS	519
READS	520
READS	521
READS	522
READS	523
READS	524
READS	525
READS	526
READS	527
READS	528
READS	529

```

4 READ(1,102)(ZC(2,I),I=1,NAP)
2 CONTINUE
100 FORMAT(4F10.0)
101 FORMAT(2F10.0,10X,F10.0)
102 FORMAT(8F10.0)
103 FORMAT(2F10.0,I2,8X,I2,9X,I1,9X,I1,9X,I1)
104 FORMAT(/)
105 FORMAT(F10.0)

      SMALL=X2
      IF(X1.LT.SMALL)SMALL=X1
11 X1=X1+CORD1
   X2=X2+CORD2
      BIG=X2
      IF(X1.GT.BIG)BIG=X1
      WIDE=Y1
      IF(Y2.GT.WIDE)WIDE=Y2
      RETURN
      END

```

C

APPENDIX C

SOURCE LISTING OF PROGRAM PLOTS

C	PROGRAM PLOTS(INPUT,OUTPUT,TAPE2=OUTPUT,TAPE3,TAPE4,TAPE5)	1	PLOTS
	THIS INITIALIZES THE PLOTTING ROUTINES AND CALLS PLOTSPAN.	2	PLOTS
	CALL INITT(120)	3	PLOTS
	CALL TERM(3,4096)	4	PLOTS
	CALL CHRISZ(4)	5	PLOTS
	CALL PLOTSPAN	6	PLOTS
	CALL FINITT(0,0)	7	PLOTS
	STOP	8	PLOTS
	END	9	PLOTS
C		10	PLOTS
	SUBROUTINE PLOTSPAN	11	PLOTS
	COMMON VORS(3,500)	12	PLOTS
	COMMON /PRAMS/NVOR,NVRNPRD,NRCV,NVOR1,NVRN	13	PLOTS
	COMMON /TITL/ TITLE(6)	14	PLOTS
	COMMON /RJT/KARRAY2(7),AMYTRIX(3,3)	15	PLOTS
	INTEGER RNCV	16	PLOTS
C		17	PLOTS
C	THIS PROGRAM READS A SET OF 3 ANGLES AND PLOTS THE	18	PLOTS
C	CONFIGURATION AFTER ROTATING IT THROUGH THE INDICATED ANGLES.	19	PLOTS
C	FIRST ONE SIDE IS PLOTTED THEN IT IS REFLECTED THROUGH	20	PLOTS
C	ITS X-Z PLANE AND THE OTHER SIDE IS PLOTTED.	21	PLOTS
C	THE ROTATIONS ARE CARRIED OUT BY MATRIX MULTIPLICATION.	22	PLOTS
C	IT IS ALSO DETERMINED IF THE SUBPANELING AND CONTROL	23	PLOTS
C	POINTS ARE TO BE SHOWN.	24	PLOTS
C		25	PLOTS
	LOGICAL SUBLINE,CPJNTZ	26	PLOTS
	DATA (KARRAY2(M),M=1,7)/20,3, 3,500,3,3,3/	27	PLOTS
	DATA PIE/3.14159/	28	PLOTS
	9 REWIND 3	29	PLOTS
	REWIND 4	30	PLOTS
	REWIND 5	31	PLOTS

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APPENDIX C. - Continued

32	PLOTS	READ(3)NPAN,TITLE,TESTRTO
33	PLOTS	
34	PLOTS	HERE THE ANGLES OF ROTATION ARE READ; ROLL, PITCH, AND YAW,
35	PLOTS	ARE READ.
36	PLOTS	
37	PLOTS	CALL ERASE
38	PLOTS	WRITE(2,104)
39	PLOTS	READ *,ROLL
40	PLOTS	IF(ROLL.GT.360.)RETURN
41	PLOTS	WRITE(2,105)
42	PLOTS	READ *, PITCH
43	PLOTS	WRITE(2,106)
44	PLOTS	READ *,YAW
45	PLOTS	WRITE(2,107)
46	PLOTS	READ 112,SUBLINE
47	PLOTS	WRITE(2,108)
48	PLOTS	READ 112,CPOINTZ
49	PLOTS	104 FORMAT(" INPUT THE ROLL ANGLE FOR THE AIRCRAFT",/,
50	PLOTS	2" (DEG), >360 TO STOP.")
51	PLOTS	105 FOPMAT("PITCH ANGLE")
52	PLOTS	106 FOPMAT("YAW ANGLE")
53	PLOTS	107 FOPMAT("SUBPANELING ? TRUE OR FALSE")
54	PLOTS	108 FOPMAT("CONTROL POINTS ? TRUE OR FALSE")
55	PLOTS	112 FOPMAT(L7)
56	PLOTS	
57	PLOTS	IF(TESTRTO.LT.1.00)GO TO 500
58	PLOTS	IF(TESTRTO.GT.1.25)TESTRTO=1.25
59	PLOTS	RYY=5.0/TESTRTO
60	PLOTS	IXX2=INT(TESTRTO*3200)
61	PLOTS	IXX1=INT((4096-IXX2)*.5)
62	PLOTS	CALL SWINDO(IXX1,IXX2,1,3200)
63	PLOTS	CALL DWINDO(-5.0,5.0,-RYY,RYY)
64	PLOTS	GO TO 501
65	PLOTS	500 CALL SWINDO(450,3200,1,3200)
66	PLOTS	CALL DWINDO(-5.0,5.0,-5.0,5.0)
67	PLOTS	501 CALL TURNIT(ROLL,PITCH,YAW)
68	PLOTS	3 CALL MOVABS(100,3500)
69	PLOTS	CALL ANMODE

APPENDIX C. - Continued

```

768 WRITE(2,788)TITLE,ROLL,PITCH,YAW
      FORMAT(8A10,/, " THE ROLL ANGLE IS ",F7.1,
1/, " THE PITCH ANGLE IS ",F7.1,
2/, " THE YAW ANGLE IS ",F7.1)
      DO 300 NPANS=1,NPAN
      DO 299 L=1,2
      READ(3)NVR,NVCV,PDL,IQUANT,NVRNPRD,ISETS
      ISET=ISETS
      NVORI=NVR+1
      NVRN=NVRNPRD-NVORI+1
      IF(SUBLINE.A.PDL.LT.360)CALL VORSUB(5HLINES,ISETS,3,L)
      IF(.NOT.SUBLINE.A.PDL.GT.360)CALL VORSUB(5HLINES,ISETS,3,L)
      IF(SUBLINE.A.PDL.GT.360)CALL RYDLIN(L)
      IF(.NOT.SUBLINE.A.PDL.LT.360)CALL VORSUB(5HEDGES,ISETS,3,L)
      IF(PDL.GF.O.O.A.CPONTZ)CALL CPOINTS(L)
      IF(SUBLINE.A.PDL.GT.360.)ISET=0
      IF(L.EQ.2.OR.IQUANT.EQ.1)GO TO 300
      CALL BACKJUP(3,ISET+1)
299 CONTINUE
300 IF(L.EQ.2.A.ISET.EQ.0.OR.IQUANT.EQ.1.A.ISET.EQ.0)CALL SKPFILE(3,
      ISETS)
      CALL MOVABS(100,3500)
      CALL BIGER(II)
      IF(II-1)9,9,3
      END
C
SUBROUTINE VORSUB(WHICH,ISETS,IFILE,L)
COMMON /PRAMS/NVR,NVRNPRD,RNCV,NVORI,NVRN
COMMON VORS(3,500)
C
C THIS SUBROUTINE PLOTS A FLAT MAJOR PANEL WITH
C CR WITHOUT SUBPANELING, OR A CURVED PANEL WITHOUT SUBPANELING.
C
CALL SAVE(IFUG,IFILE,IDUM,1,L,ISETS)
IF(WHICH.EQ.5HLINES)GO TO 1
N1=NVRN-1

```

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PLOTS	70
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PLOTS	94
PLOTS	95
PLOTS	96
PLOTS	97
PLOTS	98
PLOTS	99
PLOTS	100
PLOTS	101
PLOTS	102
PLOTS	103
PLOTS	104
PLOTS	105

APPENDIX C. - Continued

```

N2=NVOR
GO TO 2
1 N1=NVOR1
  N2=1
2 CALL SAVE(IFUG,IFILE,NVRNPRD,2,L,ISETS)
  DO 257 I=1,NVRN,N1
    DO 257 J=1,NVORI
      I2=I+J-1+IFUG
      IF(I2.LE.500)GO TO 3
10 I2=I2-500
      CALL SAVE(IFUG,IFILE,NVRNPRD,2,L,ISETS)
      IF(I2.GT.500)GO TO 10
3 IF(J.EQ.1)CALL MOVEA(VORS(1,I2),VORS(3,I2))
257 CALL DRAWA(VORS(1,I2),VORS(3,I2))
  DO 258 J=1,NVORI,N2
    IF(ISETS.EQ.1)GO TO 4
      CALL BACKUP(IFILE,ISETS)
      CALL SAVE(IFUG,IFILE,NVRNPRD,1,L,ISETS)
      CALL SAVE(IFUG,IFILE,NVRNPRD,2,L,ISETS)
4 DO 258 I=1,NVOR1,NVORI
    I2=I+J-1+IFUG
    IF(I2.LE.500)GO TO 5
      CALL SAVE(IFUG,IFILE,NVRNPRD,2,L,ISETS)
      I2=I2-500
5 IF(I.EQ.1)CALL MOVEA(VORS(1,I2),VORS(3,I2))
258 CALL DRAWA(VORS(1,I2),VORS(3,I2))
RETURN
END
C
SUBROUTINE REFLECT(N)
COMMON VORS(3,500)
THIS SUBROUTINE REFLECTS THE CONFIGURATION ACCROSS ITS X-Z PLANE.
DO 1 J=1,N
1 VORS(2,J)=-VORS(2,J)
RETURN
END
C

```

PLOTS 106  
PLOTS 107  
PLOTS 108  
PLOTS 109  
PLOTS 110  
PLOTS 111  
PLOTS 112  
PLOTS 113  
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PLOTS 140  
PLOTS 141



APPENDIX C. - Continued

178 PLOTS  
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 213 PLOTS

```

C
C
SUBROUTINE CPPOINTS(L)
  THIS SUBROUTINE PLOTS X'S ON THE CONTROL POINTS.
  COMMON VORS(3,500)
  COMMON/PRAMS/NVOR,NVRNPRD,RNCV,NVORI,NVRN
  INTEGER PRD,RNCV
  IF(L.EQ.2)CALL BACKUP(4,ISETS+1)
  READ(4)NVOR,RNCV,PRD,NNN,NVRNPRD,NNO,ISETS
  CALL SAVE(IFUG,4,IDUM,1,L,ISETS)
  CALL SAVE(IFUG,4,NNN,2,L,ISETS)
  DO 1 J=1,NNO,NVOR
  DO 1 I=1,NVOR
  K=(I+J-2)*4+IFUG
  K4=K+4
  CALL MOVEA(VORS(1,K+1),VORS(3,K+1))
  CALL DRAWA(VORS(1,K+2),VORS(3,K+2))
  CALL MOVEA(VORS(1,K+3),VORS(3,K+3))
  CALL DRAWA(VORS(1,K+4),VORS(3,K+4))
  1 IF(K4.EQ.500)CALL SAVE(IFUG,4,NNN,2,L,ISETS)
  RETURN
  END
    
```

```

C
C
SUBROUTINE PNDLINE(L)
  THIS DEFINES A SET OF PARAMETERS SO A CURVED MAJOR
  PANEL WITH SUBPANELING CAN BE PLOTTED.
  INTEGER RNCV
  COMMON VORS(3,500)
  COMMON/PRAMS/NVOR,NVRNPRD,RNCV,NVORI,NVRN
  IF(L.EQ.2)CALL BACKUP(5,ISETS+1)
  READ(5)NVOR,RNCV,NVRNPRD,ISETS
  NVORI=NVOR+1
  NVRN=NVRNPRD-1+IVOR
  CALL VORSUB(5,LINES,ISETS,5,L)
  RETURN
  END
    
```

APPENDIX C. - Continued

<pre> SUBROUTINE SAVE(FUG,FILE,P,TEST,L,SETS) THIS SUBROUTINE MAKES IT POSSIBLE TO REDUCE THE CORE REQUIREMENTS OF THE PROGRAM. INTEGER FUG,FILE,P,TEST,SETS COMMON VORS(3,500) COMMON /PRAMS/NVOR,NVRNPRD,RNCV,NVORI,NVRN COMMON/ROT/KARRAY2(7),AMYTRIX(3,3) IF(TEST.EQ.1)GO TO 1 FUG=-500*TEST2 IF(TEST2.EQ.SETS)RETURN TEST2=TEST2+1 IF(TEST2.LT.SETS)N=500 IF(TEST2.EQ.SETS)N=MOD(P,500) IF(N.EQ.0)N=500 READ(FILE)((VORS(I,J),I=1,3),J=1,N) IF(L.EQ.2)CALL REFLECT(N) KARRAY2(4)=N CALL MATOPS(KARRAY2,AMYTRIX,VORS,VORS) RETURN 1 TEST2=FUG=0 RETURN END </pre>	<p>214 PLOTS  215 PLOTS  216 PLOTS  217 PLOTS  218 PLOTS  219 PLOTS  220 PLOTS  221 PLOTS  222 PLOTS  223 PLOTS  224 PLOTS  225 PLOTS  226 PLOTS  227 PLOTS  228 PLOTS  229 PLOTS  230 PLOTS  231 PLOTS  232 PLOTS  233 PLOTS  234 PLOTS  235 PLOTS  236 PLOTS</p>
<pre> SUBROUTINE BACKUP(I,N) CAUSES FILE I TO BE BACKSPACED N RECORDS. DO 1 J=1,N 1 BACKSPACE I RETURN END </pre>	<p>237 PLOTS  238 PLOTS  239 PLOTS  240 PLOTS  241 PLOTS  242 PLOTS  243 PLOTS</p>

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APPENDIX C. - Continued

```

C
SUBROUTINE SKPFILE(I,N)
  CAUSES FILE I TO BE ADVANCED N RECORDS.
  DO 1 J=1,N
    1 READ(I)
  RETURN
END
PLOTS 244
PLOTS 245
PLOTS 246
PLOTS 247
PLOTS 248
PLOTS 249

C
SUBROUTINE TURNIT(RO,PO,YO)
C
C   THIS SUBROUTINE DETERMINES THE MAYTRIX WHICH WILL ROTATE
C   THE AIRCRAFT THROUGH THE ANGLES ROLL PITCH AND YAW AS
C   THEY ARE NORMALLY DETERMINED.
C
  DIMENSION ZAX(3,3),XAX(3,3),R1(3,3),R2(3,3),VECTOR(3)
  DIMENSION KMULT(7),MULTV(7)
  COMMON /ROT/ XARRAY2(7),AMYTRIX(3,3)
  DATA(KMULT(M),M=1,7)/20,6*(3)/
  DATA(MULTV(M),M=1,7)/20,3,3,1,3,3,3/
  DATA PIE/3.14159/
  DATA XAX(1,2),XAX(1,3),XAX(2,1),XAX(3,1),XAX(3,1)/4*(0.0)/
  DATA ZAX(1,3),ZAX(2,3),ZAX(3,1),ZAX(3,2)/4*(0.0)/
  DATA R1(1,2),R1(1,3),R1(2,1),R1(3,1)/4*(0.0)/
  DATA R2(1,2),R2(2,1),R2(2,3),R2(3,2)/4*(0.0)/
  DATA XAX(1,1),ZAX(3,3),R1(1,1),R2(2,2)/4*(1.0)/
  1 AMYTRIX(1,1)=AMYTRIX(2,2)=AMYTRIX(3,3)=1.0
  AMYTRIX(1,2)=AMYTRIX(1,3)=AMYTRIX(2,1)=AMYTRIX(2,3)=0.0
  AMYTRIX(3,1)=AMYTRIX(3,2)=0.0
  VECTOR(1)=VECTOR(2)=0.0
  VECTOR(3)=1.0
  R=RO*PIE/180.
  P=-PO*PIE/180.
  Y=-YO*PIE/180.
  RTO=PIE*.5-R
  XAX(2,2)=XAX(3,3)=COS(R)
  SINE=SIN(R)
  XAX(2,3)=SINE
  XAX(3,2)=-SINE
PLOTS 250
PLOTS 251
PLOTS 252
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PLOTS 278
PLOTS 279

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APPENDIX C. - Continued

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 315 PLOTS

```

ZAX(1,1)=ZAX(2,2)=COS(P)
SINE=SIN(P)
ZAX(1,2)=-SINE
ZAX(2,1)=SINE
R1(2,2)=R1(3,3)=COS(RTO)
SINE=SIN(RTO)
R1(2,3)=SINE
R1(3,2)=-SINE
CALL MATOPS(KMULT,XAX,AMYTRIX,AMYTRIX)
CALL MATOPS(KMULT,R1,AMYTRIX,AMYTRIX)
CALL MATOPS(KMULT,ZAX,AMYTRIX,AMYTRIX)
      TRANSPOSE MAYTRIX R1.
R1(2,3)=-SINE
R1(3,2)=SINE
CALL MATOPS(KMULTI,R1,AMYTRIX,AMYTRIX)
ZAX(1,1)=ZAX(2,2)=COS(Y)
SINE=SIN(Y)
ZAX(1,2)=-SINE
ZAX(2,1)=SINE
CALL MATOPS(MULTV,AMYTRIX,VECTOR,VECTOR)
A=VECTOR(1)
B=VECTOR(2)
C=VECTOR(3)
V=SQRT(B*B+C*C)
BOVRV=B/V
R1(2,2)=R1(3,3)=C/V
R1(2,3)=-BOVRV
R1(3,2)=BOVRV
R2(1,1)=R2(3,3)=V
R2(1,3)=-A
R2(3,1)=A
CALL MATOPS(KMULT,R1,AMYTRIX,AMYTRIX)
CALL MATOPS(KMULT,R2,AMYTRIX,AMYTRIX)
CALL MATOPS(KMULT,ZAX,AMYTRIX,AMYTRIX)
      TRANSPOSE MAYTRIX R1 AND MAYTRIX R2.
R1(2,3)=BOVRV
    
```

C

5

C

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APPENDIX C. - Concluded

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322

PLOTS  
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PLOTS  
PLOTS  
PLOTS  
PLOTS  
PLOTS

```
R1(3,2)=-BOVRV  
R2(1,3)=+A  
R2(3,1)=-A  
CALL MATOPS(KMULT,R2,AMYTRIX,AMYTRIX)  
CALL MATOPS(KMULT,R1,AMYTRIX,AMYTRIX)  
RETURN  
END
```

AST 100 - 6/8/75 PER COORD SHEET 164(REVISED), 712 LB/SEC ENGINES (AST-JP-2)

THE ROLL ANGLE IS 0.0  
THE PITCH ANGLE IS 0.0  
THE YAW ANGLE IS 0.0

FOR ENLARGEMENT INPUT YES



(a) With panel camber displayed.



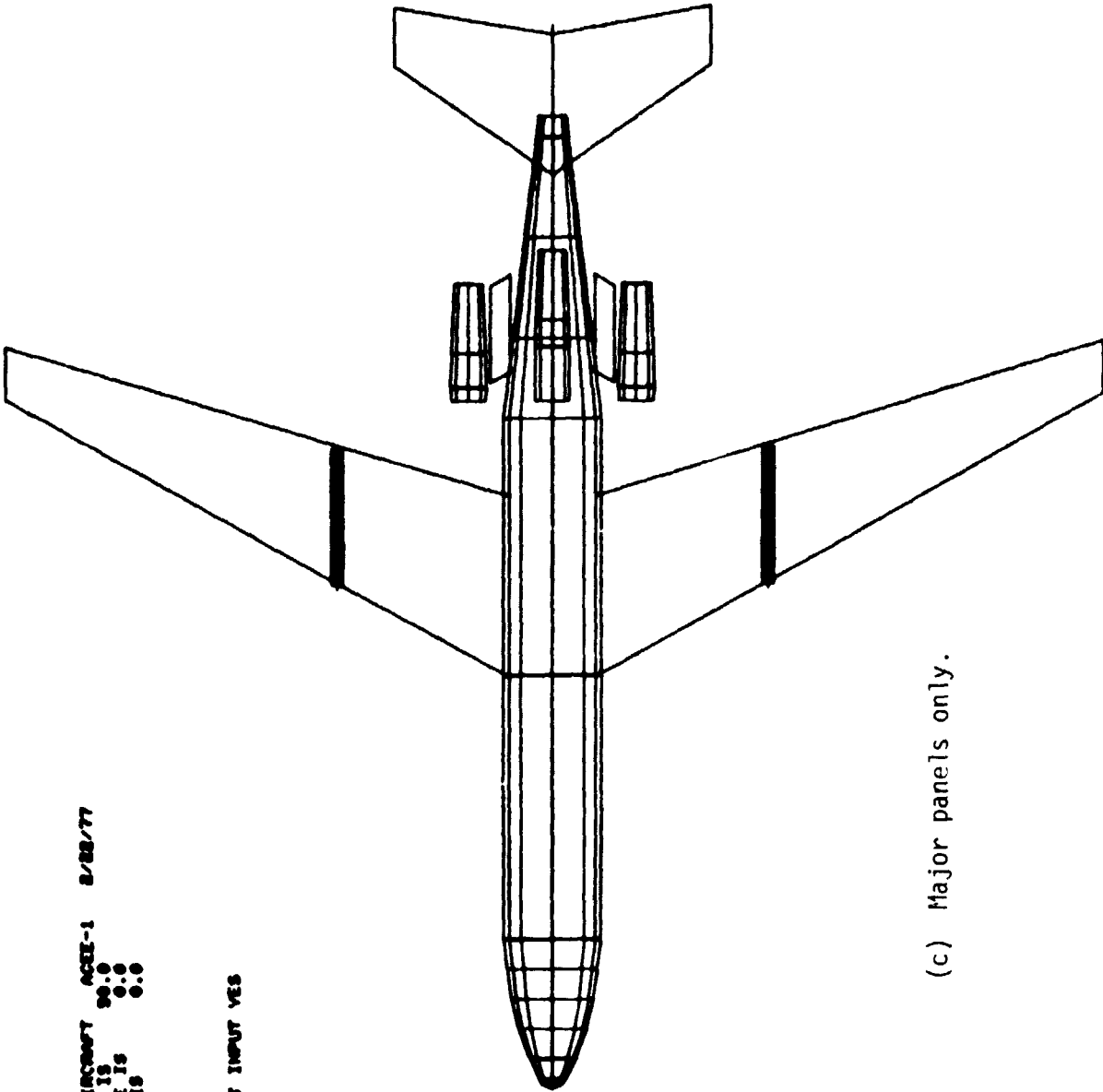
(b) Without panel camber displayed.

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Figure 1. - Display of aircraft configuration.

LAMINAR FLOW AIRCRAFT ACEE-1 8/23/77  
THE SOLL ANGLE IS 90.0  
THE PITCH ANGLE IS 0.0  
THE YAW ANGLE IS 0.0

FOR ENLARGEMENT INPUT YES

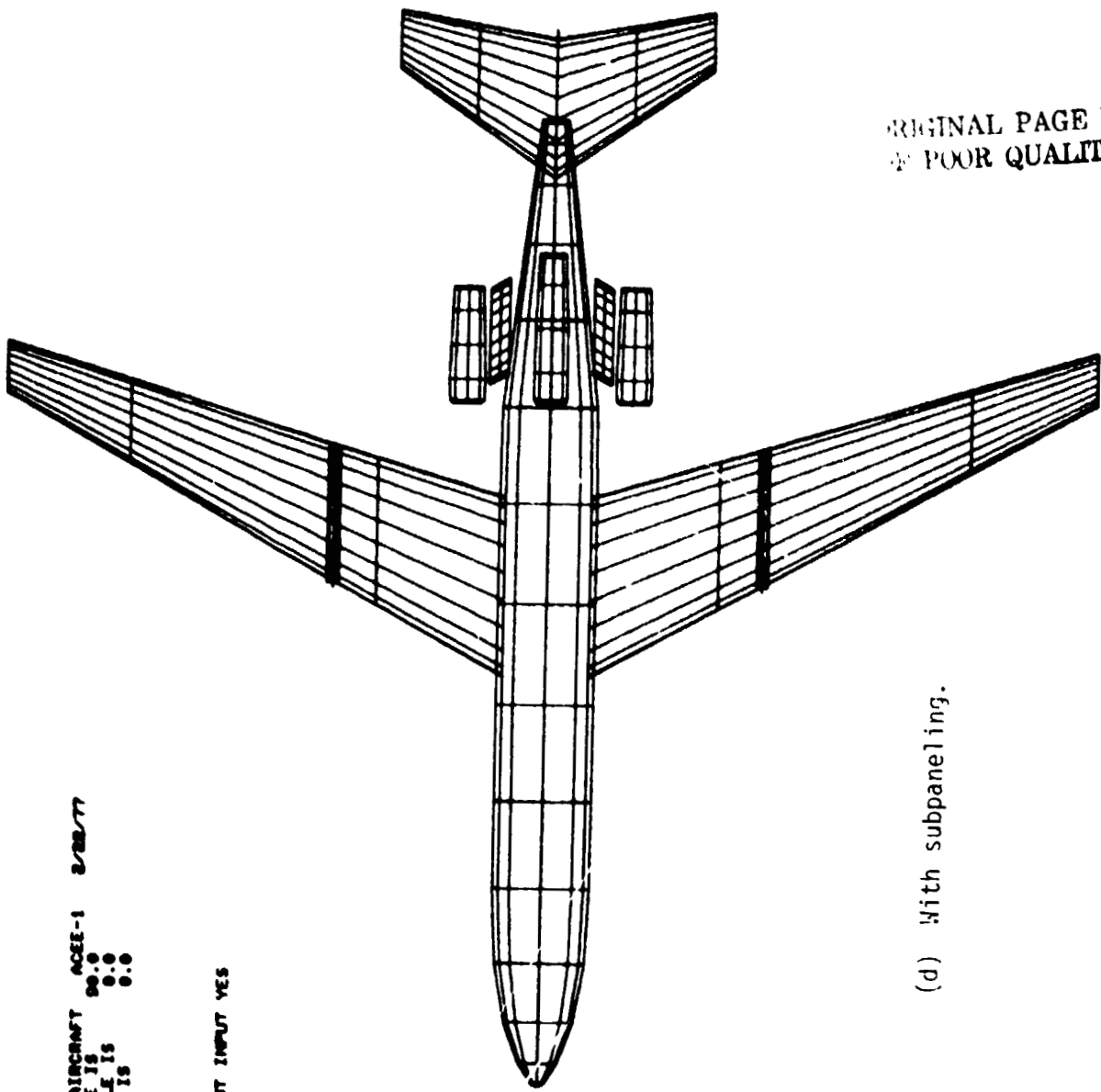


(c) Major panels only.

Figure 1. - Continued.

LAMINAR FLOW AIRCRAFT ACEE-1 2/22/77  
THE ROLL ANGLE IS 0.0  
THE PITCH ANGLE IS 0.0  
THE YAW ANGLE IS 0.0

FOR ENLARGEMENT INPUT YES



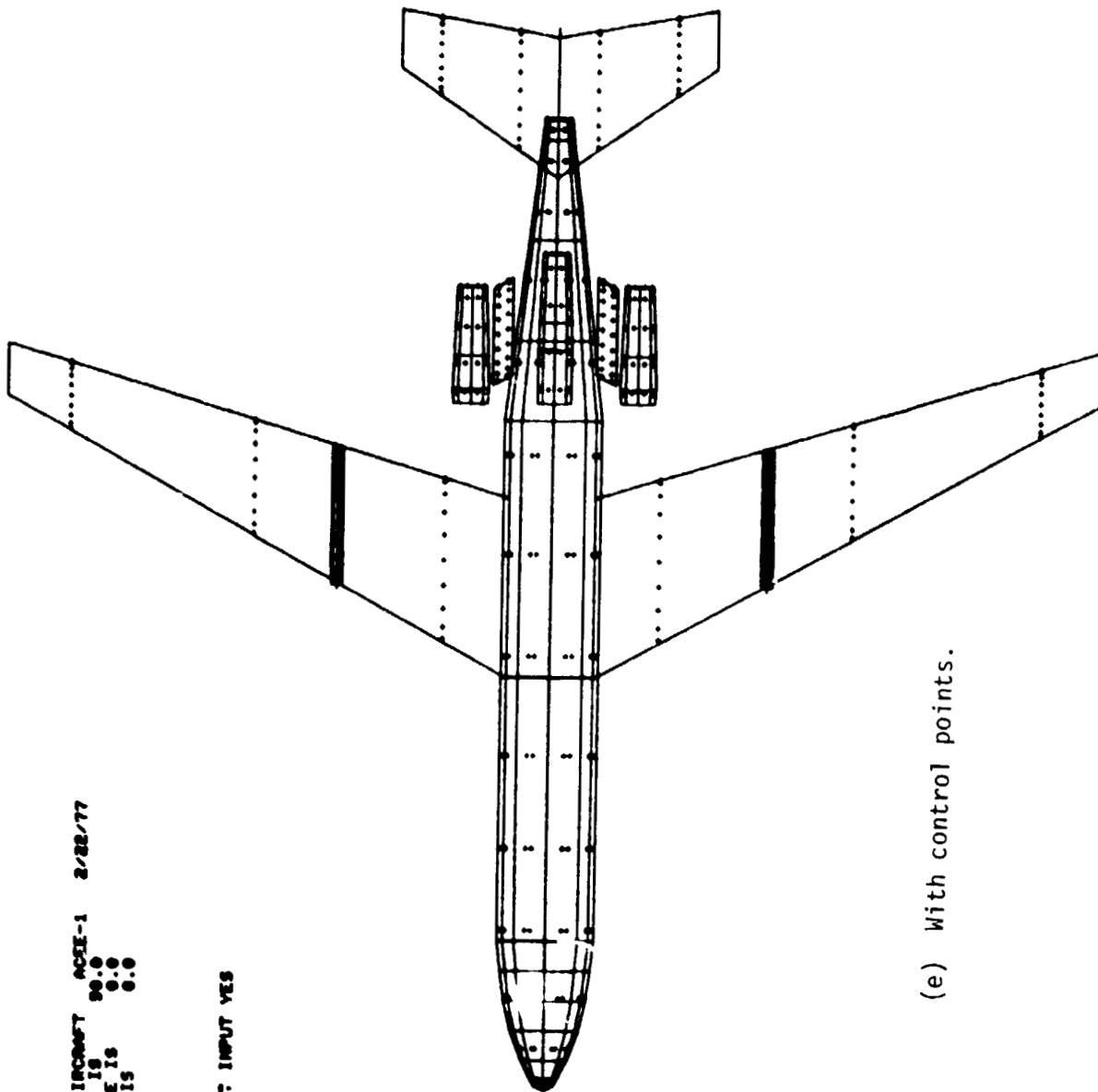
ORIGINAL PAGE IS  
OF POOR QUALITY

(d) With subpaneling.

Figure 1. - Continued.

LAMINAR FLOW AIRCRAFT ACEE-1 2/22/77  
THE ROLL ANGLE IS 00.0  
THE PITCH ANGLE IS 00.0  
THE YAW ANGLE IS 00.0

FOR ENLARGEMENT: INPUT YES

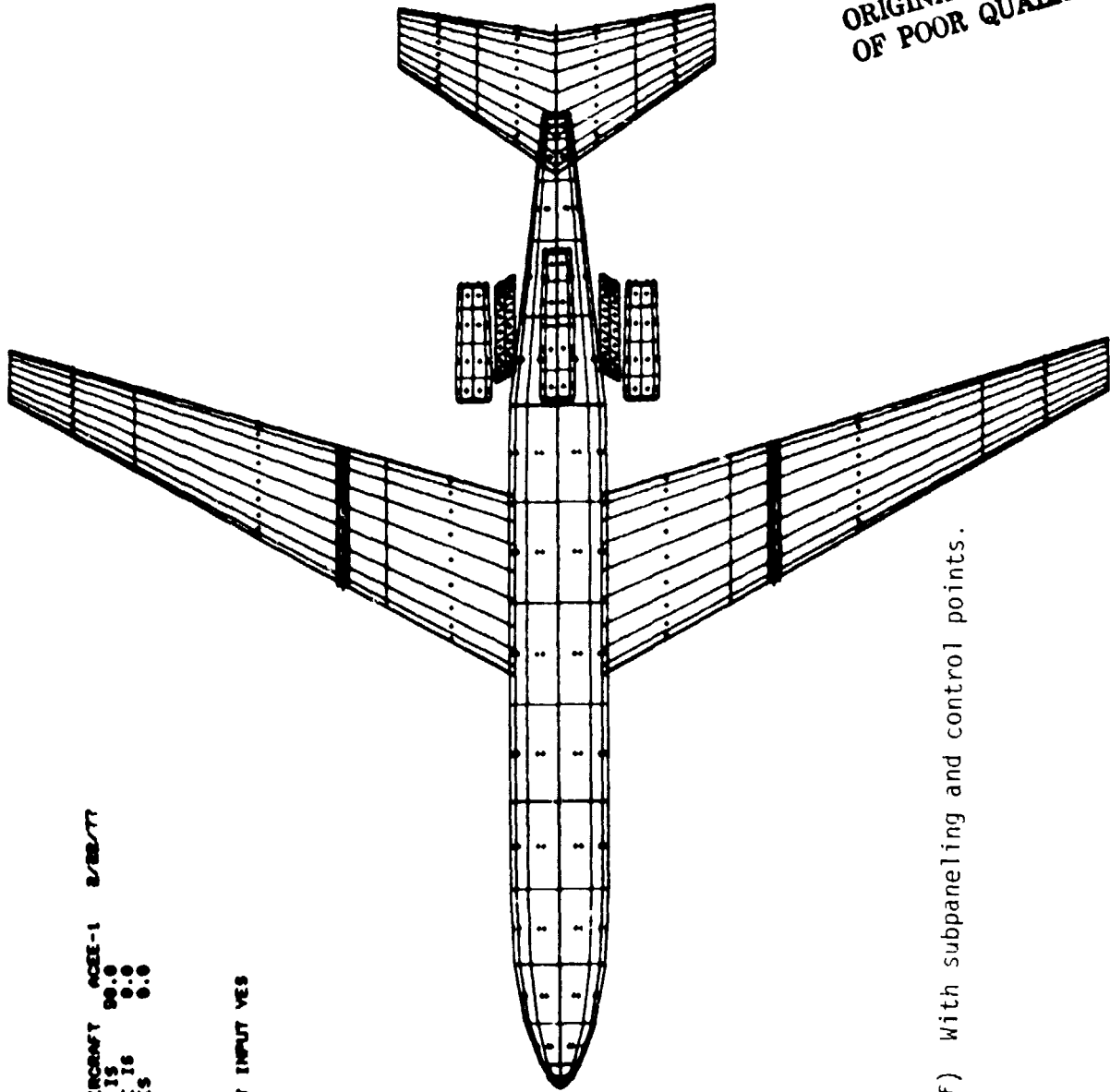


(e) With control points.

Figure 1. - Continued.

LANTING FLOW AIRCRAFT ACEE-1 2/28/77  
THE ROLL ANGLE IS 90.0  
THE PITCH ANGLE IS 0.0  
THE YAW ANGLE IS 0.0

FOR ENLARGEMENT INPUT YES



ORIGINAL PAGE IS  
OF POOR QUALITY

(f) With subpaneling and control points.

Figure 1. - Concluded.

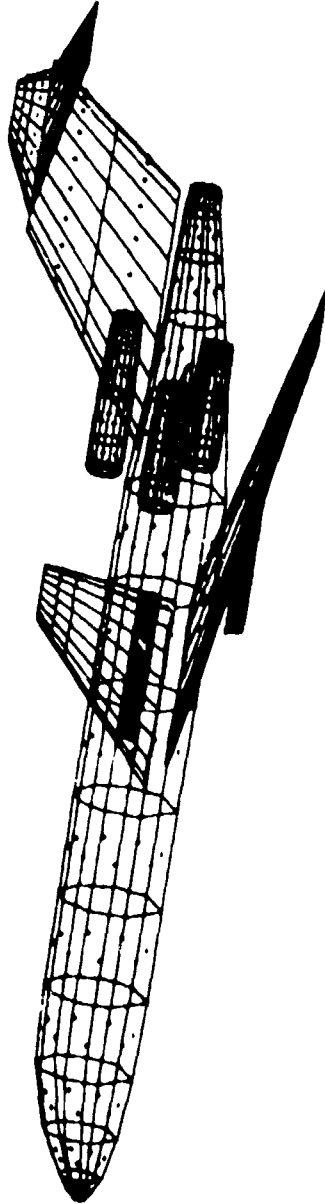
```
/GET,PLOTIT/UN=214737C
/CALL,PLOTIT(T-UORLAXX)
DO YOU WANT CAMBER IN THE PANELS ?
TRUE OR FALSE
? TRUE

    INPUT THE ROLL ANGLE FOR THE AIRCRAFT
    (DEG), >360 TO STOP.
? 11
PITCH ANGLE
? 12
YAW ANGLE
? 13
SUBPANELING ? TRUE OR FALSE
? TRUE
CONTROL POINTS ? TRUE OR FALSE
? TRUE
```

Figure 2. - Sample execution of program PLOTIT.

LAMINAR FLOW AIRCRAFT ACEE-1 2/28/77  
THE ROLL ANGLE IS 11.0  
THE PITCH ANGLE IS 12.0  
THE YAW ANGLE IS 13.0

FOR ENLARGEMENT INPUT YES

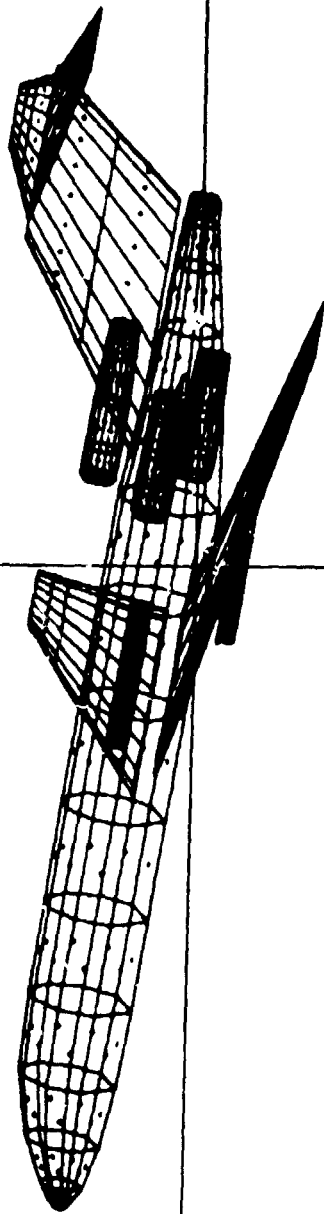


ORIGINAL PAGE IS  
OF POOR QUALITY

Figure 3. - Plot from sample execution in figure 2.

LAUNCHER FLOW AIRCRAFT ACCE-1 8/28/77  
THE ROLL ANGLE IS 11.0  
THE PITCH ANGLE IS 18.0  
THE YAW ANGLE IS 13.0

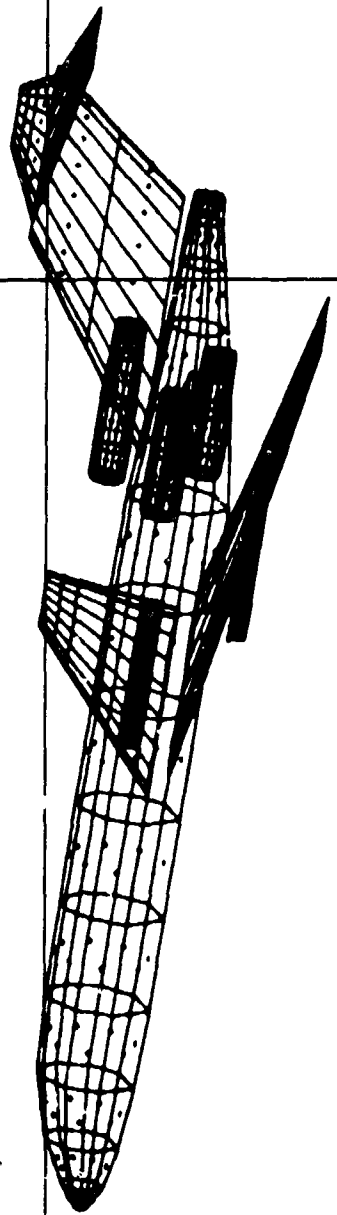
FOR ENLARGEMENT INPUT YES



(a) First location of graphics cursor.  
Figure 4. - Use of graphics cursor to enlarge a section of a configuration plot.

LANTIER FLOW AIRCRAFT ACEE-1 8/28/77  
THE ROLL ANGLE IS 11.0  
THE PITCH ANGLE IS 18.0  
THE YAW ANGLE IS 13.0

FOR C...MENT INPUT YES



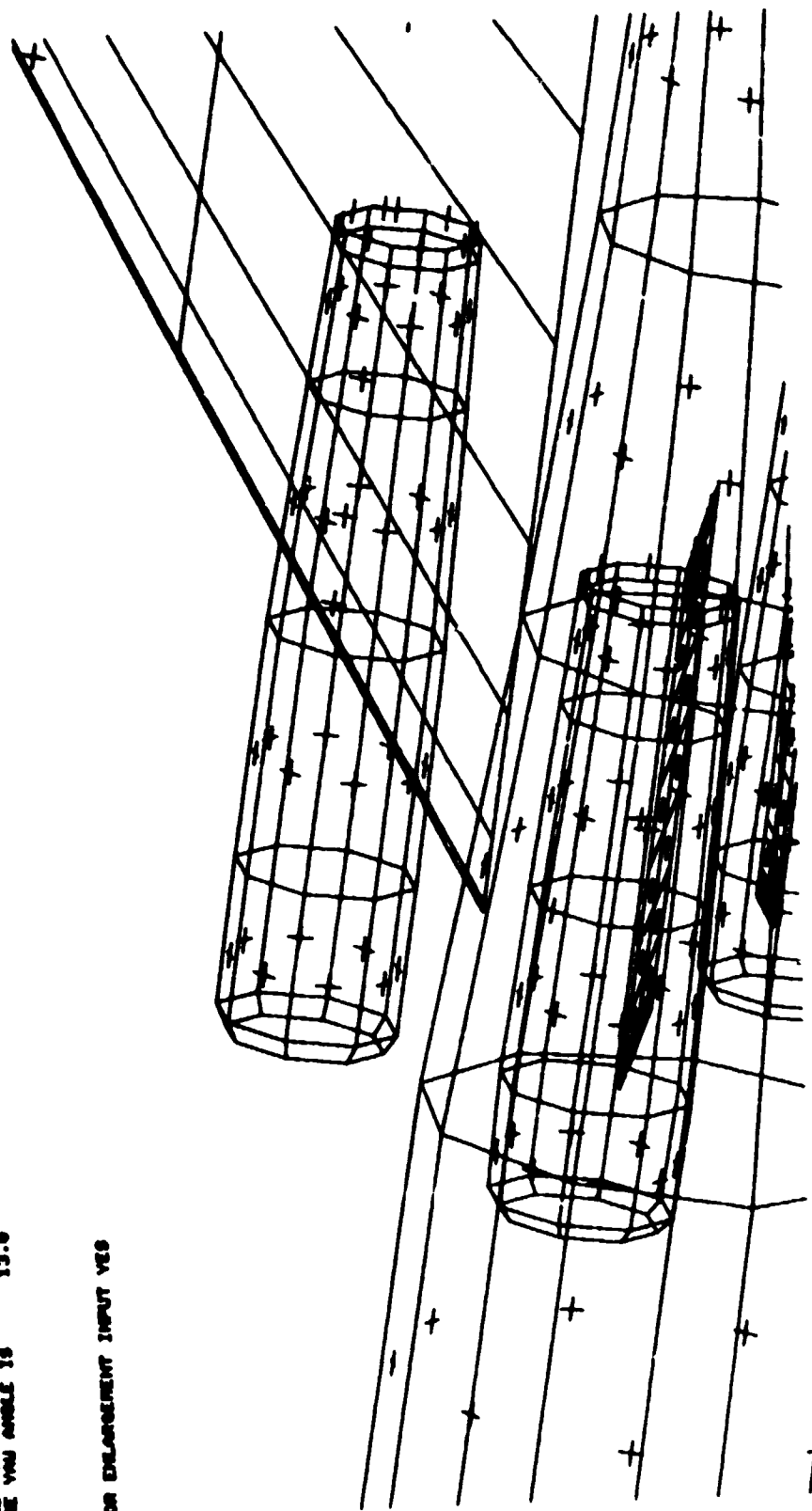
ORIGINAL PAGE IS  
OF POOR QUALITY

(b) Second location of graphics cursor.

Figure 4. - Continued.

LASTING FLOW AIRCRAFT ACES-1 8/25/77  
THE ROLL ANGLE IS 11.0  
THE PITCH ANGLE IS 18.0  
THE YAW ANGLE IS 13.0

FOR ENLARGEMENT INPUT YES



(c) Enlarged plot resulting from cursor locations in figure 4 part (a) and part (b).

Figure 4. - Concluded.