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MODE PROPAGATION IN NONUNIFORM CIRCULAR DUCTS WITH POTENTIAL FLOW†

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Abstract

A previously reported closed form solution is expanded to determine effects of isentropic mean flow on mode propagation in a slowly converging-diverging duct - a circular cosh duct. On the assumption of uniform steady fluid density, the mean flow increases the power transmission coefficient. The increase is directly related to the increase of the cutoff ratio at the duct throat. With the negligible transverse gradients of the steady fluid variables, the conversion from one mode to another is negligible, and the power transmission coefficient remains unchanged with the mean flow direction reversed. With a proper choice of frequency parameter, many different modes can be made subject to a single value of the power transmission loss. The paper also describes a systematic method to include the effects of the gradients of the steady fluid variables.

Nomenclature

$A(x), \tilde{A}(x)$	functions, Eqs. (61), (83)
a	effective length of converging-diverging section
$B_m(x), B_0(x)$	functions determining duct shape with or without mean flow, Eqs. (45), (48)
$b(x)$	duct radius varying along duct
b_0	duct radius at throat ($x = 0$)
b_-	radius of left hand side uniform duct (inlet)
b_+	radius of right hand side uniform duct (exit)
c	constant sound speed in isochoric flow
c_0	sound speed in still fluid
$D_1, D_2, D_c, D_s, D_r, D_t$	constants
F	hypergeometric function
G	Green's function

g_{mn}	one-dimensional Green's function
h	radial variable in cylindrical coordinate system
\vec{I}	acoustic power flux
J_m	Bessel function of order m
K	total acoustic power across duct
k	free space wave number ω/c_0
k_- or $k_{mn}^{(-)}$	propagation constant of a mode or (m,n) mode in inlet
$k_+, k_{mn}^{(+)}$	propagation constant of a mode or (m,n) mode in exit
L_p, L_ξ, L_x, L_s	linear operations
$M(x)$	Mach number of mean flow of constant steady density fluid
M_0	Mach number at duct throat
M_-	Mach number of uniform flow in inlet
M_+	Mach number of uniform flow in exit
m, m'	circumferential mode numbers
n, n'	radial mode numbers
\hat{n}	unit vector normal to surface
P	steady pressure in flow
P_c	steady pressure in flow of constant density
P_δ	deviation $P - P_c$
p	acoustic pressure
Q_{mn}	normalized Bessel function corresponding to (m,n) mode, Eq. (72)
q	parameter, Eq. (56)
R	acoustic power reflection coefficient
$R(x)$ or $R(r)$	axial wave function, Eqs. (30), (40), (41)
r	radial variable in local spherical coordinate system ($dr \approx dx$)
$S(\vec{x}, t)$	interaction term, Eq. (17)

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$S_\omega(\vec{x})$	temporal Fourier transform of $S(x,t)$
T	acoustic power transmission coefficient
t	time
TL	acoustic power transmission loss, Eq. (118)
\vec{U}	steady velocity of isentropic flow
\vec{U}_C	steady velocity of constant density fluid flow
\vec{U}_δ	deviation $\vec{U} - \vec{U}_C$
\vec{u}	acoustic velocity field
v	a constant, Eq. (57)
w	a parameter, Eqs. (45) and (46)
w_0	w with $M = 0$
X 's	see Eqs. (105) to (109)
x	axial coordinate
\vec{x}, \vec{x}_0	three dimensional coordinates
z	dimensionless axial coordinate, Eq. (55)
α or α_{mn}	eigenvalue corresponding to a duct mode or (m,n) mode
$\beta(x)$	$b(x)/b_0$
β_\pm	b_\pm/b_0
Γ	Gamma function
γ	specific heat ratio
γ_0	mode cut-off ratio at duct throat, Eq. (120)
γ_-	mode cut-off ratio at inlet, Eq. (119)
$\delta(x-x_0)$	Dirac delta function
$\delta_{n,n'}$	Kronecher's delta
ϵ	a parameter, 1 for $M = 0$, Eq. (45)
ζ	a frequency parameter, Eqs. (53) and (122)
η	normalized coordinate variable, Eq. (32)
θ	polar angle in local spherical coordinate system
θ_0	half cone angle of duct segment, Fig. 1
μ	a parameter, Eq. (47)
ν	frequency parameter, kb_0/α

ξ	see Eq. (12)
ρ	acoustic perturbation of density
ρ_0	spatially varying steady density
ρ_C	constant steady density
ρ_δ	deviation $\rho_0 - \rho_C$
σ	a parameter, Eq. (66)
τ	b_+/b_-
$\Phi(\vec{x})$	a Fourier transform of acoustic velocity potential
ϕ	azimuthal angle variable
x	see Eq. (13)
$\Psi(x,t)$	acoustic velocity potential
ψ	axial wave function, Eq. (41)
ω	angular frequency

Introduction

In a previously reported investigation of mode propagation in a nonuniform duct,¹ a closed form solution was obtained for a particular class of converging-diverging ducts - circular cosh ducts. These ducts are composed of two asymptotically uniform ducts which are smoothly coupled through a converging-diverging section. The two uniform duct elements can differ from each other in cross sectional area. The duct shape can be adjusted by means of three duct parameters to produce a variety of circular ducts of practical interest. The analysis, which was previously developed for the case of no mean flow, is here expanded to include mean flow effects.

The mean flow under consideration is subsonic, and is assumed to be isentropic. Viscous effects are excluded from discussion. Numerical results will be presented only for cases of low Mach numbers, for which the steady density and pressure hardly vary from one location to another in the duct. However a systematic method is presented to treat finite deviations of the steady fluid variables from the flow of constant density.

Wave Equation

As customary, the fluid variables are first decomposed into steady and fluctuating parts. The steady part will be further decomposed to treat its spatial variation. The steady density, pressure and flow velocity are denoted respectively by ρ_0 , P and U ; and the fluctuations by ρ , p and u . Retaining only the first order terms of the fluctuations, one obtains from the continuity equation

$$\left(\frac{\partial}{\partial t} + \vec{U} \cdot \vec{\nabla}\right)p + \gamma P \vec{\nabla} \cdot \vec{u} + \frac{\gamma P}{\rho_0} \vec{\nabla} \rho_0 \cdot \vec{u} + (\vec{\nabla} \cdot \vec{U})p = 0. \quad (1)$$

And from the Euler's equation

$$\left(\frac{\partial}{\partial t} + \vec{U} \cdot \vec{\nabla}\right) \vec{U} + (\vec{U} \cdot \vec{\nabla}) \vec{U} + \frac{1}{\rho_0} \vec{\nabla} p - \frac{\vec{\nabla} p}{\gamma \rho_0} p = 0. \quad (2)$$

Here γ is the specific heat ratio of the fluid, and we have used the adiabatic relation between density and pressure

$$p = \frac{\gamma P}{\rho_0} \rho. \quad (3)$$

Also used are the zeroth order relations

$$\vec{\nabla} \cdot (\rho_0 \vec{U}) = 0, \quad (4)$$

$$\rho_0 (\vec{U} \cdot \vec{\nabla}) \vec{U} + \vec{\nabla} p = 0 \quad (5)$$

The steady variables are, in general, functions of spatial coordinates in a nonuniform duct with mean flow. The functions may be complicated. However, in a subsonic flow, the major portion of the steady variables may assume simple coordinate dependence, and the deviations from the simple dependence can be small. We now decompose the steady variables further as follows:

$$\rho_0 = \rho_c + \rho_\delta, \quad (6)$$

$$p = p_c + p_\delta, \quad (7)$$

$$\vec{U} = \vec{U}_c + \vec{U}_\delta. \quad (8)$$

Here ρ_c , p_c and \vec{U}_c are a new set of steady variables which are simple functions of spatial coordinates, and ρ_δ , p_δ and \vec{U}_δ are the deviations. In the present analysis, ρ_c and p_c are chosen to be constant and set equal to the steady density and pressure in the left side uniform duct element ($x < 0$, or the inlet). The variable \vec{U}_c is the velocity of the constant density flow, and therefore, it is subject to the isochoric flow condition

$$\vec{\nabla} \cdot \vec{U}_c = 0 \quad (9)$$

On inserting Eqs. (6) to (8) into Eqs. (1) and (2), one obtains

$$\left(\frac{\partial}{\partial t} + \vec{U}_c \cdot \vec{\nabla}\right) p + \gamma p_c \vec{\nabla} \cdot \vec{U} + \xi = 0, \quad (10)$$

$$\left(\frac{\partial}{\partial t} + \vec{U}_c \cdot \vec{\nabla}\right) \vec{U} + (\vec{U} \cdot \vec{\nabla}) \vec{U}_c + \frac{1}{\rho_c} \vec{\nabla} p + \vec{\nabla} \xi = 0, \quad (11)$$

where

$$\xi = \vec{\nabla} \cdot (p \vec{U}_\delta) + \gamma p_\delta \vec{\nabla} \cdot \vec{U} + \vec{U} \cdot \vec{\nabla} p_\delta, \quad (12)$$

$$x = \vec{U}_\delta \cdot \vec{U} - \frac{\rho_\delta}{\rho_c} \left(1 - \frac{\rho_\delta}{\rho_c}\right) p + O(\rho_\delta^3 p). \quad (13)$$

With the potential flow condition $\vec{\nabla} \times \vec{U}$, one sets

$$\vec{U} = \vec{\nabla} \Psi(\vec{x}, t), \quad (14)$$

where Ψ is the acoustic velocity potential. With the substitution of this equation, Eqs. (10) and (11) are combined to yield the equation

$$\nabla^2 \Psi - \frac{1}{c^2} \left(\frac{\partial}{\partial t} + \vec{U}_c \cdot \vec{\nabla}\right)^2 \Psi = S(\vec{x}, t) \quad (15)$$

where

$$c^2 = \frac{\gamma p_c}{\rho_c} = c_0^2 - \frac{\gamma - 1}{2} U_c^2 (x = -\infty) \quad (16)$$

$$S = \frac{1}{c^2} \left\{ \left(\frac{\partial}{\partial t} + \vec{U}_c \cdot \vec{\nabla}\right) x - \frac{\xi}{\rho_c} \right\}. \quad (17)$$

Equation (15) is, strictly speaking, a homogeneous equation for the acoustic field. It is, however, convenient to treat it as an inhomogeneous wave equation with the interaction term S . With $S = 0$, Eq. (15) is the homogeneous wave equation in a mean flow of constant density and pressure. The term S represents the interaction between the acoustic field and the mean flow field due to the spatial variations ρ_δ , p_δ and \vec{U}_δ of the latter. Since no acoustic energy is created or annihilated, the interaction accounts for the elastic scattering of sound by the spatial non-uniformity of the steady fluid variables.

Equation (15) is linear for the acoustic field. Thus the temporal Fourier transform yields

$$\nabla^2 \phi(\vec{x}) + \frac{1}{c^2} (\omega + i \vec{U}_c \cdot \vec{\nabla})^2 \phi(\vec{x}) = S_\omega(\vec{x}) \quad (18)$$

Here

$$\phi(\vec{x}) = \frac{1}{\sqrt{2\pi}} \int_{-\infty}^{\infty} \Psi(\vec{x}, t) e^{i\omega t} dt. \quad (19)$$

$$S_\omega(\vec{x}) = \frac{1}{c^2} \left\{ (-i\omega + \vec{U}_c \cdot \vec{\nabla}) x_\omega(\vec{x}) + \frac{1}{\rho_c} \xi_\omega(\vec{x}) \right\}, \quad (20)$$

with

$$x_\omega(\vec{x}) = \frac{1}{\sqrt{2\pi}} \int_{-\infty}^{\infty} x(\vec{x}, t) e^{i\omega t} dt, \quad (21)$$

$$\xi_\omega(\vec{x}) = \frac{1}{\sqrt{2\pi}} \int_{-\infty}^{\infty} \xi(\vec{x}, t) e^{i\omega t} dt. \quad (22)$$

Homogeneous Solutions

We consider solutions to Eq. (18) first with the interaction term neglected. The wave equation is then

$$\nabla^2 \phi_H + \frac{1}{c^2} (\omega + i\vec{U}_C \cdot \vec{\nabla})^2 \phi_H = 0 \quad (23)$$

where the subscript H has been used to indicate the homogeneous solutions. The homogeneous solutions serve two purposes: First, with boundary conditions properly satisfied, they describe mode propagation in the duct with mean flow of uniform density, which can be adequately described by the variables ρ_C , p_C and \vec{U}_C . It is likely that potential flow solutions will be close to these variables for low Mach numbers. Secondly, the homogeneous solutions can be used as basis functions to construct the Green's function. With the help of the latter, Eq. (18) can be systematically solved by means of the iteration-perturbation method for cases when the deviations of steady fluid variables are finite but small.²

Let us consider Eq. (23) in a circular duct which is specified in terms of the radius $b(x)$, x being the axial coordinate variable. As in Ref. 1, we consider first a small duct segment and the local spherical coordinates (r, θ, ϕ) as illustrated in Fig. 1. The azimuth ϕ , the angle variable around the duct axis, is not shown in the figure. The duct segment is so short that it may be regarded locally conical. Thus the slope, $b = db/dx$, of the duct wall remains constant within the segment. The origin of the local spherical coordinate system is located at a point on the duct axis such that the coordinate surface $\theta = \theta_0$ (cone) tangentially contacts the duct wall in the segment. The half cone angle θ_0 is related to the wall slope b' as

$$b' = \tan \theta_0. \quad (24)$$

With the spherical coordinates, the isochoric flow velocity is chosen as

$$\vec{U}_C = \hat{r} \frac{\Omega}{r^2} \quad (25)$$

where Ω is a constant with the sign being opposite for converging and diverging sections, and \hat{r} is the radial unit vector. Equation (23) can then be written for the spherical coordinates as

$$\frac{1}{r^2} \left\{ \frac{\partial}{\partial r} r^2 \frac{\partial}{\partial r} + \frac{1}{\sin \theta} \frac{\partial}{\partial \theta} \sin \theta \frac{\partial}{\partial \theta} + \frac{1}{\sin^2 \theta} \frac{\partial^2}{\partial \phi^2} \right\} \phi_H + \frac{1}{c^2} \left(\omega + \frac{i\Omega}{r^2} \frac{\partial}{\partial r} \right)^2 \phi_H = 0 \quad (26)$$

This equation can be separated into three ordinary differential equations. With the substitution

$$\phi_H(\vec{x}) = R(r) J(\theta) H(\phi) \quad (27)$$

one obtains, from Eq. (26)

$$\left(\frac{d^2}{d\phi^2} + m^2 \right) H = 0, \quad (28)$$

$$\frac{1}{\sin \theta} \frac{d}{d\theta} \sin \theta \frac{dJ}{d\theta} + \left(\frac{\alpha^2}{\sin^2 \theta_0} - \frac{m^2}{\sin^2 \theta} \right) J = 0, \quad (29)$$

$$(1 - M^2) \frac{d^2 R}{dr^2} + \left[\frac{2(1 + M^2)}{r} - 2ikM \right] \frac{dR}{dr} + \left[k^2 - \left(\frac{\alpha}{r \sin \theta_0} \right)^2 \right] R = 0. \quad (30)$$

where m and α are the separation constants, and

$$k = \frac{\omega}{c}, \quad M = \frac{\Omega}{cr^2}.$$

The solution to Eq. (28) is

$$H = e^{im\phi} \quad (31)$$

with m being an integer. As for Eq. (29), it is convenient to introduce the normalized coordinate variable n , defined as

$$n = \frac{\sin \theta}{\sin \theta_0} \quad \text{for } \theta_0 \neq 0$$

$$= \frac{h}{b} \quad \text{for } \theta_0 = 0. \quad (32)$$

where h is the radial variable in the cylindrical coordinate system. The value of n varies from 0 to 1 in the duct. On the duct axis, $n = 0$; and on the duct wall, $n = 1$. Equation (30) is written in terms of n as

$$\left(1 - \sin^2 \theta_0 n^2 \right) \frac{d^2 J}{dn^2} + \frac{1}{n} \left(1 - 2 \sin^2 \theta_0 n^2 \right) \frac{dJ}{dn} + \left(\alpha^2 - \frac{m^2}{n^2} \right) J = 0. \quad (33)$$

For a duct of slowly varying cross section ($|b'| \ll 1$), $\sin \theta_0 \approx b'$. Neglecting terms involving $(b')^2$ and higher orders, one can write Eq. (33) as

$$\frac{d^2 J}{dn^2} + \frac{1}{n} \frac{dJ}{dn} + \left(\alpha^2 - \frac{m^2}{n^2} \right) J = 0. \quad (34)$$

This is the Bessel equation of order m , and the physically acceptable solution is

$$J \sim J_m(\alpha n) \quad (35)$$

where J_m is the Bessel function of order m . From the boundary condition of the hard duct wall, one determines values of α as follows:

$$\alpha = \alpha_{mn}, n = 0, 1, 2, \dots \quad (36)$$

where α_{mn} is the n -th zero of $dJ_m(x)/dx$.

Some remarks should be made on the coordinate variable n . Unlike the variable θ , n is not a local coordinate variable, but it is valid through the entire duct. To see this, let us consider two small duct segments 1 and 2, respectively at $x = x_1$ and x_2 . The half cone angles of the respective segments are denoted by $\theta_{0,1}$ and $\theta_{0,2}$. Assume $\theta_{0,1} > \theta_{0,2}$. The variable θ varies from 0 to $\theta_{0,1}$ in segment 1, and from 0 to $\theta_{0,2}$ in segment 2. A value of θ greater than $\theta_{0,2}$ and less than $\theta_{0,1}$, represents a coordinate surface in segment 1, but not in segment 2. Furthermore, two coordinate surfaces represented by a value of θ less than $\theta_{0,2}$, respectively in the two duct segments are not, in general, equivalent to each other. On the other hand, n varies from 0 to 1 for all the duct segments. The locations represented by a value of n are in one-to-one correspondence for two different duct segments. A coordinate surface is uniquely determined as the locus of $n = n_1$, n_1 being a constant, and the coordinate surface spans all the duct segments. A coordinate surface with a value of n does not intersect with any other coordinate surface with a different value of n . The duct wall is a coordinate surface corresponding to $n = 1$. We also note, from Eqs. (35) and (36), that an eigenfunction assumes the same value on a coordinate surface $n = n_1$. A mode is in one-to-one correspondence with an eigenfunction. Thus, with the use of the variable n , a mode is defined in a nonuniform circular duct with isochoric flow if $|b'|$ remains small (Fig. 2).

Equation (30) governs the propagation of a mode along the duct. The equation is first transformed from the local coordinate r to the duct coordinate x . To this end we use the relations

$$\sin \theta_0 = b' \quad (37)$$

$$r = \frac{b}{b'} \quad (38)$$

$$\frac{d}{dr} = \frac{d}{dx} \quad (39)$$

These relations are valid for all the duct segments and the error is of order $(b')^2$. In order to avoid misunderstandings, a remark should be made on Eq. (39): Eq. (30) governs the wave along a line that is the intersection of coordinate surfaces $\phi = \text{constant}$ and $n = \text{constant}$ (not

$\theta = \text{constant}$). On this line, $dr = dx \sqrt{1 + b'^2}$, and Eq. (39) follows. One should not attempt to derive a similar relation by regarding r as a function of h as well as x . Such a derivation might lead one to an incorrect conclusion, and the error would appear to be of order b' instead of $(b')^2$. Note that the origin of the local coordinate system changes with x . It cannot be overemphasized that R (as given in Eq. (30)) deals with the wave field variation along a line which is the intersection of coordinate surfaces $\phi = \text{constant}$ and $n = \text{constant}$. Finally, the use of the relations (37), (38), and (39) is very unlikely to lead one to any asymptotic paradox.

Using Eqs. (37) to (39), one can write Eq. (30) as

$$(1 - M^2) \frac{d^2 R}{dx^2} + \left[\frac{2b'}{b} (1 + M^2) + 2ikM \right] \frac{dR}{dx} + \left(k^2 - \frac{\alpha^2}{b^2} \right) R = 0. \quad (40)$$

Here we have used

$$M(x) = \frac{M_0}{\beta^2(x)}$$

$$\beta(x) = \frac{b(x)}{b_0}$$

where b_0 and M_0 are the duct radius and the Mach number at the duct throat. For a given $\beta(x)$, $M(x)$ is determined once M_0 is known. Thus, for a given duct shape, M_0 is sufficient to specify $M(x)$.

With the substitution

$$R(x) = \frac{\psi(x)}{b(x) \sqrt{1 - M^2(x)}} \exp \left(-ik \int^x \frac{M(x)}{1 - M^2(x)} dx \right), \quad (41)$$

Equation (40) is transformed into

$$\frac{d^2 \psi}{dx^2} + \left[\left(\frac{k}{1 - M^2} \right)^2 - \frac{1}{1 - M^2} \frac{\alpha^2}{b^2} \right] \psi = 0, \quad (42)$$

where terms like b'^2 and b'' have been neglected.

Solutions in a Circular Cosh Duct

Equation (42) was solved exactly in a closed form with a circular cosh duct when $M = 0$, in Ref. 1. With $M \neq 0$, it is convenient to introduce a function $B_M(x)$ as follows:

$$B_M(x) = \frac{1}{(1 - M^2)\beta^2} - v^2 \frac{M^2(2 - M^2)}{(1 - M^2)^2}, \quad (43)$$

where $v = kb_0/\alpha$. Note that this function depends on the Mach number and the frequency (v) as well as the duct shape ($\beta(x)$). With the use of Eq. (43), (42) is written as

$$\frac{d^2 \psi}{dx^2} + \left[k^2 - \frac{\alpha^2}{b_0^2} B_M(x) \right] \psi = 0. \quad (44)$$

The discussions thus far apply to any nonuniform hardwall circular duct as long as b'^2 and b'' are negligible.

To utilize the cosh duct solution, we set

$$B_M(x) = B_M(0) + w \left\{ \cosh^2 \mu \operatorname{sech}^2(\epsilon x/a - \mu) - \sinh(2\mu) \tanh(\epsilon x/a - \mu) - \cosh(2\mu) \right\}. \quad (45)$$

Here a is the effective length of the converging-diverging section, and w , μ and ϵ are parameters depending on M , v and duct geometry. The parameters w and μ are determined as

$$w = \left\{ [B_M(0) - B_M(\infty)][B_M(0) - B_M(-\infty)] \right\}^{1/2}, \quad (46)$$

$$\mu = \frac{1}{4} \ln \left\{ \frac{B_M(0) - B_M(+\infty)}{B_M(0) - B_M(-\infty)} \right\}. \quad (47)$$

Involved in these equations are M , v and β_{\pm} the radius ratios of the duct. For fixed values of M_0 and v , w and μ are determined from the radius ratios β_{\pm} , and vice versa. One notices from Eq. (47) that μ is equal to zero for a symmetric duct ($\beta_+ = \beta_-$), regardless of values of M_0 and v .

The parameter ϵ is not given in a closed form, but its value can be determined analytically. For fixed values of M_0 and v , the duct radius ratios β_{\pm} have been used for the determination of w and μ . The axial variation of the radius in the converging-diverging section will be used to determine ϵ . If the value of ϵ were kept constant, the duct shape would depend on M and v as can be readily seen from Eqs. (43) and (45). Thus, the value of ϵ must be so determined that the duct shape may remain unchanged for different values of M and v . To this end, at a properly chosen axial location in the converging-diverging section, the duct radius is required to be the same for all the allowed values of M_0 and v . We choose $M_0 = 0$ as the reference case. With $M_0 = 0$, we set $\epsilon = 1$, and obtain from Eq. (43)

$$B_0(x) \equiv B_M(x) \Big|_{M=0} = \beta^{-2}(x), \quad (48)$$

and from Eq. (45)

$$B_0(x) = 1 + w_0 \left\{ \cosh^2 \mu_0 \operatorname{sech}^2(x/a - \mu_0) - \sinh(2\mu_0) \tanh(x/a - \mu_0) - \cosh(2\mu_0) \right\} \quad (49)$$

where the subscript 0 indicates $M = 0$. Note that B_0 , w_0 and μ_0 do not depend on v . The axial location is chosen preferably in the converging section, as $x = -s$. The duct radius at that location is determined from Eqs. (48) and (49):

$$\beta(-s) = \left\{ 1 + w_0 \cosh^2 \mu_0 \operatorname{sech}^2(s/a + \mu_0) - \sinh(2\mu_0) \tanh(s/a - \mu_0) - \cosh(2\mu_0) \right\}^{1/2} \quad (50)$$

With $M_0 \neq 0$, Eq. (43) is written for $x = -s$ as

$$B_M(-s) = \frac{1}{[1 - M^2(-s)]\beta^2(-s)} - v^2 \frac{M^2(-s)[2 - M^2(-s)]}{[1 - M^2(-s)]^2} \quad (51)$$

and Eq. (45) is

$$B_M(-s) = B_M(0) + w \left\{ \cosh^2 \mu \operatorname{sech}^2(\epsilon s/a + \mu) + \sinh(2\mu) \tanh(\epsilon s/a + \mu) - \cosh(2\mu) \right\} \quad (52)$$

The value of ϵ is determined from Eqs. (50) to (52). There may exist more than one real solution for ϵ ; the correct one should be determined from physical considerations. A list of values of ϵ are given in Table 1, for various values of M_0 and ζ , where ζ is a frequency parameter defined as

$$\zeta = \frac{1}{1 - M_0^2} \left(kb_0 - \alpha \sqrt{1 - M_0^2} \right) \quad (53)$$

As the table shows, ϵ is close to unity in most cases of interest. Also notice that ϵ takes the same value for all the modes at $\zeta = 0$. The value of ϵ at $\zeta = 0$ are listed for various values of M_0 and β_- in Table 2.

We now return to Eq. (44). On inserting Eq. (45) into Eq. (44) we obtain

$$\frac{d^2 \psi}{dz^2} + \left[q^2 - v \left(\cosh^2 \mu \operatorname{sech}^2 z - \sinh(2\mu) \tanh z \right) \right] \psi = 0 \quad (54)$$

where

$$z = \epsilon x/a - \mu \quad (55)$$

$$q^2 = (ka/\epsilon)^2 - (\alpha a/\epsilon b_0)^2 [B_M(0) - w \cosh(2\mu)]. \quad (56)$$

$$v = (\alpha a/\epsilon b_0)^2 w \quad (57)$$

Equation (54) can be solved exactly.² The general solution is written as

$$\psi = D_1 \psi_1(z) + D_2 \psi_2(z) \quad (58)$$

Here D_1 and D_2 are constants; and ψ_1 and ψ_2 are linearly independent solutions.

$$\psi_1 = A(z) F(\zeta_1, \zeta_2, \zeta_3; (1 + e^{2z})^{-1}), \quad (59)$$

$$\psi_2 = A(z) (1 + e^{2z})^{-ik_+ a/\epsilon}$$

$$F(\zeta_2 - \zeta_3 + 1, \zeta_1 - \zeta_3 + 1, 2 - \zeta_3; (1 + e^{2z})^{-1}), \quad (60)$$

where F is the hypergeometric function, and

$$A(z) = e^{i(k_+ - k_-)za/2\epsilon} (2 \cosh z)^{i(k_+ + k_-)a/2\epsilon}, \quad (61)$$

$$k_{\pm} = \frac{k}{1 - M_{\pm}^2} \left[1 - (1 - M_{\pm}^2) \left(\frac{\alpha}{kb_{\pm}} \right)^2 \right]^{1/2}, \quad (62)$$

$$b_{\pm} = b(x = \pm\infty), \quad M_{\pm} = M(x = \pm\infty),$$

$$\zeta_1 = \left\{ 1 - i[(k_+ + k_-)a/\epsilon + \sigma] \right\} / 2 \quad (63)$$

$$\zeta_2 = \left\{ 1 - i[(k_+ + k_-)a/\epsilon - \sigma] \right\} / 2 \quad (64)$$

$$\zeta_3 = 1 - ik_+a/\epsilon \quad (65)$$

$$\sigma = (4v \cosh^2 \mu - 1)^{1/2} \quad (66)$$

The constants D_1 and D_2 are to be determined from boundary conditions at $x = \pm\infty$. The asymptotic forms of the hypergeometric function are used to find

i) at $x = +\infty$

$$\psi_1 = e^{-ik_+ua/\epsilon} e^{ik_+x}, \quad (67)$$

$$\psi_2 = e^{ik_+ua/\epsilon} e^{-ik_+x}, \quad (68)$$

ii) at $x = -\infty$

$$\psi_1 = \frac{\Gamma(\zeta_3)\Gamma(\zeta_1 + \zeta_2 - \zeta_3)}{\Gamma(\zeta_1)\Gamma(\zeta_2)} e^{-ik_+ua/\epsilon + ik_-x} + \frac{\Gamma(\zeta_3)\Gamma(\zeta_3 - \zeta_1 - \zeta_2)}{\Gamma(\zeta_3 - \zeta_1)\Gamma(\zeta_3 - \zeta_2)} e^{ik_+ua/\epsilon - ik_-x}, \quad (69)$$

$$\psi_2 = \frac{\Gamma(2 - \zeta_3)\Gamma(\zeta_1 + \zeta_2 - \zeta_3)}{\Gamma(\zeta_2 - \zeta_3 + 1)\Gamma(\zeta_1 - \zeta_3 + 1)} e^{-ik_+ua/\epsilon + ik_-x} + \frac{\Gamma(2 - \zeta_3)\Gamma(\zeta_3 - \zeta_1 - \zeta_2)}{\Gamma(1 - \zeta_1)\Gamma(1 - \zeta_2)} e^{ik_+ua/\epsilon - ik_-x}. \quad (70)$$

where Γ is the Gamma function.

Mode Propagation in Circular Cosh Ducts With Low Mach Number Flow

As mentioned earlier, the deviations ρ_δ , P_δ and U_δ are negligible in cases of low Mach numbers ($M_0 < 0.5$). In these cases, the solutions can be obtained as in Eq. (58), with the boundary conditions to be satisfied. Note that these solutions are the zeroth order solutions to Eq. (18) and can be used to obtain more accurate solutions.

Let us begin with the incident wave that is composed of the (m, n) mode coming from $x = -\infty$. In the incident side uniform duct ($x \ll 0$),

$$\phi_1 = Q_{mn}(n) e^{im\phi} \frac{1}{b_- \sqrt{1 - M_-^2}} \times \exp \left[i \left(-kM_-/(1 - M_-^2) + k_{mn}^{(-)} \right) x \right], \quad (71)$$

where

$$Q_{mn}(n) = J_m(\alpha_{mn}n) / L_{mn} \quad (72)$$

$$L_{mn} = \left\{ \int_0^1 J_m^2(\alpha_{mn}n) n \, dn \right\}^{1/2} \quad (73)$$

$$k_{mn}^{(\pm)} = k_{\pm} \Big|_{\alpha = \alpha_{mn}} \quad (74)$$

Correspondingly, we have

$$\psi_1 = e^{ik_{mn}^{(-)}x}. \quad (75)$$

With mode conversion neglected, the solution ψ is given at $x \pm \infty$ in the form

$$\lim_{x \rightarrow -\infty} \psi(x) = e^{ik_{mn}^{(-)}x} + D_r e^{-ik_{mn}^{(-)}x}, \quad (76)$$

$$\lim_{x \rightarrow \infty} \psi(x) = D_t e^{ik_{mn}^{(+)}x} \quad (77)$$

The second term of Eq. (76) represents the reflected wave, and Eq. (77) the transmitted wave. The constants in Eq. (58) are readily determined by comparing the asymptotic expressions in Eqs. (67) to (70) with Eqs. (76) and (77):

$$D_2 = 0 \quad (78)$$

$$D_1 = e^{ik_{mn}^{(-)}ua/\epsilon} \frac{\Gamma(\zeta_1)\Gamma(\zeta_2)}{\Gamma(\zeta_3)\Gamma(\zeta_1 + \zeta_2 - \zeta_3)} \quad (79)$$

The coefficients D_r and D_t are

$$D_r = e^{2ik_{mn}^{(-)} \mu a / \epsilon} \frac{\Gamma(\zeta_1) \Gamma(\zeta_2) \Gamma(\zeta_3 - \zeta_1 - \zeta_2)}{\Gamma(\zeta_1 + \zeta_2 - \zeta_3) \Gamma(\zeta_3 - \zeta_1) \Gamma(\zeta_3 - \zeta_2)}, \quad (80)$$

$$D_t = e^{i(k_{mn}^{(-)} - k_{mn}^{(+)}) \mu a / \epsilon} \frac{\Gamma(\zeta_1) \Gamma(\zeta_2)}{\Gamma(\zeta_3) \Gamma(\zeta_1 + \zeta_2 - \zeta_3)} \quad (81)$$

In Eqs. (78) to (81) the parameters μ , ϵ , ζ_1 , ζ_2 and ζ_3 contain the constant α : it is to be replaced by the eigenvalue α_{mn} . The solution is

$$\phi = Q_{mn}(n) e^{im\phi} b^{-1} (1 - M^2)^{-1/2} \times \exp \left(ik \int_0^x \frac{M}{1 - M^2} dx \right) \tilde{A}(z) \Gamma(\zeta_1, \zeta_2, \zeta_3; (1 + e^{2z})^{-1}) \quad (82)$$

where

$$\tilde{A}(z) = D_1 A(z) \exp \left(i \frac{kM_-}{1 - M^2} x_- + ik \int_{x_-}^0 \frac{M}{1 - M^2} dx \right) \quad (83)$$

x_- being a large negative value of x . Recall that $n = h/b(x)$ and $z = \epsilon x/a - \mu$.

Green's Function and General Solutions

When the interaction term S_w is finite but small, Eq. (18) can be solved by means of the iteration-perturbation method.³ To this end, the equation is first replaced by the integral equation

$$\phi(\vec{x}) = \phi^{(0)}(\vec{x}) + \int G(\vec{x}, \vec{x}_0) S_w(\vec{x}_0) d^3 \vec{x}_0 \quad (84)$$

Here G is the Green's function to be discussed, and the function $\phi^{(0)}$ has been used for the homogeneous solution, to let the solution satisfy the boundary conditions.

The equation for the Green's function is

$$\left\{ \nabla^2 + \frac{1}{2} \left(\omega + i \vec{U}_c \cdot \vec{\nabla} \right)^2 \right\} G(\vec{x}, \vec{x}_0) = \delta(\vec{x} - \vec{x}_0) \quad (85)$$

The solution can be obtained in terms of the homogeneous solutions with proper boundary conditions as follows:

$$G(\vec{x}, \vec{x}_0) = \frac{1}{2\pi} \sum_m e^{im(\phi - \phi_0)} \times \sum_n Q_{mn}(n) Q_{mn}(n_0) g_{mn}(x, x_0) \quad (86)$$

Here g_{mn} is the one-dimensional Green's function. On inserting Eq. (86) into Eq. (85) and using the orthogonality and the completeness of the eigenfunctions, one obtains

$$\left\{ (1 - M^2) \frac{d^2}{dx^2} + \left[\frac{2b'}{b} (1 + M^2) - 2ikM \right] \frac{d}{dx} + \left(k^2 - \frac{\alpha_{mn}^2}{b^2} \right) \right\} g_{mn}(x, x_0) = \delta(x - x_0). \quad (87)$$

The homogeneous solution of this equation is none other than R given in Eq. (41).

Consider the factor ψ first. For a circular cosh duct, the homogeneous solutions ψ_1 and ψ_2 are given in Eqs. (59) and (60). The source term is non-zero only in the converging-diverging section. Thus, the Green's function is required to represent outgoing waves in the uniform duct elements. Consider a new set of homogeneous solutions $\psi_>$ and $\psi_<$ which asymptotically behave as

$$\psi_>(x) \xrightarrow{x \rightarrow \infty} e^{ik_{mn}^{(+)} x}, \quad (88)$$

$$\psi_<(x) \xrightarrow{x \rightarrow -\infty} e^{-ik_{mn}^{(-)} x}. \quad (89)$$

Comparing these equations with Eqs. (67) to (70), one can readily find

$$\psi_> = \psi_1, \quad (90)$$

$$\psi_< = \psi_1 - \frac{\Gamma(\zeta_3) \Gamma(\zeta_2 - \zeta_3 + 1) \Gamma(\zeta_1 - \zeta_3 + 1)}{\Gamma(\zeta_1) \Gamma(\zeta_2) \Gamma(2 - \zeta_3)} \psi_2. \quad (91)$$

Corresponding to these functions, the homogeneous solutions to Eq. (87) are

$$R_>(x) = b^{-1} (1 - M^2)^{-1} \exp \left(-ik \int_0^x \frac{M}{1 - M^2} dx \right) \psi_>(x), \quad (92)$$

$$R_<(x) = b^{-1} (1 - M^2)^{-1} \exp \left(-ik \int_x^0 \frac{M}{1 - M^2} dx \right) \psi_<(x). \quad (93)$$

The one-dimensional Green's function is then obtained as

$$g_{mn}(x, x_0) = \frac{1}{x_0 \Delta(x_0)} \begin{cases} R_>(x) R_<(x_0) & \text{for } x > x_0 \\ R_<(x) R_>(x_0) & \text{for } x < x_0 \end{cases} \quad (94)$$

where Δ is the Wronskian

$$\Delta = \begin{vmatrix} R_< & R_<' \\ R_> & R_>' \end{vmatrix} \quad (95)$$

The source S_ω , as given in Eq. (20) is a linear function of acoustic field variables. The latter are not known until the problem is solved. However, when the deviations, ρ_δ , P_δ and \vec{U}_δ are small, the source is small. Thus, the zeroth order solution $\phi(0)$ can be used for the acoustic field variables to compute S_ω . The acoustic velocity is $\vec{u}(0) = \vec{v}\phi(0)$, and the acoustic pressure is obtained from Eqs. (10) and (11) with ξ and x neglected.

$$p(0) = L_p \phi(0), \quad (96)$$

where L_p is a linear operator defined as

$$L_p = i\rho_c c \left\{ \left(\frac{\alpha}{kb} \right)^2 - \left[iM + \frac{2(1+M^2)b'}{kb} \right] \frac{\partial}{\partial x} - \frac{1-M^2}{k} \frac{\partial^2}{\partial x^2} \right\}. \quad (97)$$

It follows that

$$\xi_\omega(0) = L_\xi \phi(0), \quad (98)$$

$$x_\omega(0) = L_x \phi(0) \quad (99)$$

Here L_ξ and L_x are linear operators containing the deviations of the mean fluid variables:

$$L_\xi = (\vec{v} \cdot \vec{U}_\delta + \vec{U}_\delta \cdot \vec{v}) L_p - \gamma P_\delta \left(\frac{\alpha^2}{b^2} - \frac{2b'}{b} \frac{\partial}{\partial x} - \frac{\partial^2}{\partial x^2} \right) + \vec{v} P_\delta \cdot \vec{v}, \quad (100)$$

$$L_x = \vec{U}_\delta \cdot \vec{v} - \frac{\rho_\delta}{\rho_c} \left(1 - \frac{\rho_\delta}{\rho_c} \right) L_p - \frac{1-M^2}{k} \frac{\partial^2}{\partial x^2} \quad (101)$$

The source term is then

$$S_\omega(0) = L_S \phi(0) \quad (102)$$

with the operator defined as

$$L_S = \frac{1}{c^2} \left\{ (-i\omega + U_c \frac{\partial}{\partial x}) L_x + \frac{1}{\rho_c} L_\delta \right\}. \quad (103)$$

On inserting Eqs. (86) and (102) into Eq. (84), one obtains

$$\phi(\vec{x}) = \phi(0)(\vec{x}) + \frac{1}{\sqrt{2\pi}} \times \sum_{m'n'} e^{im'\phi} Q_{m'n'}(\eta) X_{mn;m'n'}(x), \quad (104)$$

where the incident wave is the (m,n) mode (see Eq. (82)), and

$$X_{mn;m'n'}(x) = \times \frac{1}{2\pi} \int_0^{2\pi} d\phi_0 \int_0^1 \eta_0 d\eta_0 \int dx_0 e^{-im'\phi_0} Q_{m'n'}(\eta_0) \times g_{m'n'}(x, x_0) L_S e^{im'\phi_0} Q_{mn}(\eta_0) R^{(0)}(x_0). \quad (105)$$

For an axisymmetric flow, L_S is independent of ϕ_0 thus one can readily obtain

$$X_{mn;m'n'} = \delta_{nm} X_{nn}^m, \quad (106)$$

$$X_{nn}^m = \int \eta_0 d\eta_0 \int dx_0 Q_{mn}(\eta_0) g_{mn}(x, x_0) \times L_S Q_{mn}(\eta_0) R^{(0)}(x_0). \quad (107)$$

Furthermore, if ρ_δ , P_δ and U_δ are independent of η and if $\hat{n} \cdot \vec{U}_\delta = 0$, one has then

$$X_{nn}^m = \delta_{nn} X_{nn}^m(x), \quad (108)$$

$$X_{nn}^m = \int dx_0 g_{mn}(x, x_0) L_S R^{(0)}(x_0). \quad (109)$$

It follows that mode conversion, from one mode to another, occurs only if the transversal gradients of the steady fluid variables are nonnegligible.³

Power Reflection and Transmission Coefficients

The acoustic power intensity in the isentropic flow is given by⁴

$$I = \langle p \vec{u} \rangle + \vec{U} \left(\langle p^2 \rangle / \rho_0 c^2 + \vec{U} \cdot \langle p \vec{u} \rangle / c \right) + \rho_0 \langle (\vec{U} \cdot \vec{u}) \vec{u} \rangle, \quad (110)$$

where $\langle \rangle$ stands for the time average. The total power across the duct is obtained as

$$K = \int \vec{I} \cdot \hat{n} dA \quad (111)$$

where dA is the surface area element and \hat{n} the unit vector normal to the surface. In the uniform duct element containing the axisymmetric mean flow, we have

$$K = 2\pi \int I_x h dh \quad (112)$$

with the axial component I_x of the power intensity given by

$$I_x = \langle pu_x \rangle (1 + M^2) + \rho_0 U \left(\langle u_x^2 \rangle + \langle p^2 \rangle \rho_0 c^2 \right). \quad (113)$$

Let K_i , K_r and K_t denote the total powers respectively of the incident, the reflected and the transmitted wave. The power reflection and reflection coefficients, R and T , are defined as

$$R = K_r / K_i, \quad (114)$$

$$T = K_t / K_i. \quad (115)$$

Using Eqs. (71), (81), (76) and (77), one can obtain

$$R = \frac{\cosh [\pi(k_+ - k_-)a/\epsilon] + \cosh (\pi\sigma)}{\cosh [\pi(k_+ + k_-)a/\epsilon] + \cosh (\pi\sigma)} \quad (116)$$

$$T = \frac{2 \sinh (\pi k_+ a/\epsilon) \cdot \sinh (\pi k_- a/\epsilon)}{\cosh [\pi(k_+ + k_-)a/\epsilon] + \cosh (\pi\sigma)} \quad (117)$$

Note that $R + T = 1$. That is, the acoustic energy is conserved as it should be in a linear analysis of the acoustic field. The terms neglected in the approximation would not create or annihilate the acoustic energy. They, if included, might transfer the energy between modes or between the reflected and the transmitted waves. Also note that Eqs. (116) and (117) are similar to Eqs. (25) and (26) in Ref. 1. The mean flow dependence is included only through the parameters k_{\pm} , ϵ and σ . The latter are even functions of M . Thus the power reflection and transmission coefficients are independent of the mean flow direction, the positive or the negative x -direction.

Numerical results will be discussed in terms of the acoustic power transmission loss (TL) in the circular co-sh ducts. The mean flow is isentropic flow and has a uniform steady fluid density. As mentioned earlier, a uniform steady density is a good approximation for low Mach number flow in inlet ducts. Numerical calculation includes the Mach numbers up to the value of 0.5 at the duct throat.

The TL is defined as

$$TL = -10 \log_{10}(T) \quad (118)$$

For the present calculation, Eq. (117) is used for T . The TL is plotted as a function of the throat Mach number M_0 or of the frequency parameter γ_- or ζ . The parameter γ_- is the mode cut-off ratio referenced to the left hand side uniform duct element, and is referred to as the inlet cut-off ratio:

$$\gamma_- = kb_- / \alpha \sqrt{1 - M_-^2}. \quad (119)$$

The inlet cut-off ratio is a convenient parameter characterizing the modes contained in the incident wave. The sound generated in a fan duct often comprises many different modes. In such case, the mode distribution can be obtained as a function of the cut-off ratio. Another cut-off ratio useful for discussion is that referenced to the duct throat, given by

$$\gamma_0 = kb_0 / \alpha \sqrt{1 - M_0^2}. \quad (119)$$

If $\gamma_0 > 1$, the mode is cut-on through the entire duct. On the other hand, if $\gamma_0 < 1$, the mode that is cut-on initially ($\gamma_- > 1$), is cut-off in the converging-diverging section. Except for low frequencies ($kb_0 < 3$), the TL is approximately 3 dB for $\gamma_0 = 1$, independent of the duct geometry or the Mach number. The two cut-off ratios are related to each other as

$$\gamma_- = \gamma_0 \beta_- \left[(1 - M_0^2) (1 - M_0^2 \beta_-^4) \right]^{1/2} \quad (121)$$

The parameter ζ is defined in Eq. (53), and can be written as

$$\zeta = \frac{kb_0}{1 - M_0^2} \left(1 - \frac{1}{\gamma_0} \right). \quad (122)$$

This parameter collapses the TL curves for many different modes.

In Fig. 3, the TL divided by kb_- is plotted as a function of γ_- for the various values of M_0 between 0 and 0.5, and for $\beta_- = 1.1$ or 1.3. The other duct parameters are $a/b_- = 1$, $\zeta = 1$. On the top of the figure are shown the arrow marks, each pointing to a value of γ_- which corresponds to $\gamma_0 = 1$ with the given value of M_0 (cf. Eq. (121)). The (8,5) mode has been used for the calculation. However, except in the vicinity of $\gamma_0 = 1$, the results may be used for modes with eigenvalue from 8 to 25, within an error of 1 dB. As expected, the TL decreases with the increasing value of γ_- (see Ref. 1). The decreasing rate of the TL is faster for larger Mach number. For a fixed value of γ_- , the TL is smaller for larger Mach number. Since, TL/kb_- , is plotted in Fig. 4, TL itself is proportional to kb_- for a fixed value of γ_- . It follows that, for a mode distribution given as a function of the inlet cut-off ratio, the converging-diverging duct contour reduces the transmitted sound more effectively for cases when the distribution comprises modes of the larger eigenvalues.

In Fig. 4, the TL/kb_0 is plotted as a function of M_0 , for fixed values of γ_- . As M_0 increases, the TL decreases very slowly in the beginning ($M_0 \approx 0$), and then decreases at a gradually increasing rate until M_0 reaches a value for which $\gamma_0 = 1$. This mean flow dependence of the TL is directly related to the change of the cut-off ratio at the throat. As one can notice from Eq. (112), γ_0 increases with increasing M_0 for γ_- held constant.

In Fig. 5, the TL is plotted as a function of ζ . Each curve, for a value of M_0 , includes many modes with eigenvalues greater than 5. The TL increases with increasing ζ , and becomes approximately 3 dB at $\zeta = 0$ independent of the Mach number and the duct parameters. The advantage of this presentation is that almost all the modes are subject to the same TL for a given value of ζ . The mean flow dependence of the TL in this figure appears misleading. However, note that, with M_0 increased, γ_0 increases and kb_0 should be decreased to keep ζ unchanged (see Eq. (122)).

Concluding Remarks

In an attempt to improve the understanding of the acoustic characteristics of a fan duct system, mode propagation has been investigated in a particular class of converging-diverging circular ducts - circular cosh ducts, with isentropic and inviscid mean flow. The duct shape can be adjusted by means of three duct parameters, and covers a wide range of converging-diverging ducts of practical interest.

On the assumption that the duct cross section area varies slowly, an approximate wave equation has been derived. The equation is divided into two parts: (1) the homogeneous wave equation involving the mean flow of uniform steady density fluid, and (2) the interaction between the acoustic field and the deviation of the steady fluid variables from the constant steady density fluid flow. The homogeneous equation is solved in a closed form, and the interaction term is treated by means of an iteration-perturbation method. The solution of the homogeneous equation is regarded a good approximation in cases of low Mach number mean flow.

With the interaction neglected, a mode is preserved, and the acoustic power transmission coefficient increases with the mean flow increase. This mean flow effect is directly related to the change of the cut-off ratios: With the increasing mean flow, the cut-off ratio at the duct throat increases faster than at the uniform duct sections (inlet and exit). For given duct geometry and mean flow, the TL of many different modes can be made to collapse onto a single curve with a proper choice of frequency parameter (ζ). The mean flow effects remain unchanged with the flow reversed. The interaction term needs numerical computations for its detailed discussions. However, the formal solution shows that a mode may be converted to others only if the transverse gradients of the steady fluid variables are not negligible.

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TABLE 1. - VALUE OF ϵ

B_-	M_0	Eigenvalue	ζ			
			-0.4	-0.2	0	0.2
1.1	0.2	4.20		1.0020	1.0014	1.0007
		8.54	1.0020	1.0017		1.0010
		16.53	1.0017	1.0015	↓	1.0012
		23.27	1.0016	1.0015	↓	1.0012
	0.4	4.20			0.9920	0.9845
		8.54	0.9980	0.9952	↓	.9886
		16.53	.9953	.9937		.9903
		23.27	.9944	.9932	↓	.9908
1.3	0.2	4.20	1.0039	1.0026	1.0012	0.9997
		8.54	1.0025	1.0019	↓	1.0004
		16.53	1.0019	1.0015		1.0008
		23.27	1.0017	1.0014	↓	1.0009
	0.4	4.20	1.0041	0.9948	0.9839	0.9710
		8.54	.9946	.9895	↓	.9778
		16.53	.9897	.9868		.9808
		23.27	.9880	.9860	↓	.9817

TABLE 2. - VALUE OF ϵ AT $\zeta = 0$

M_0	B_-				
	1.1	1.2	1.3	1.4	1.5
0.1	1.0004	1.0006	1.0005	1.0004	1.0002
.2	1.0014	1.0015	1.0012	1.0005	.9998
.3	1.0008	1.0001	.9987	.9971	.9954
.4	.9920	.9875	.9837	.9806	.9776
.5	.9438	.9316	.9271	.9244	.9223

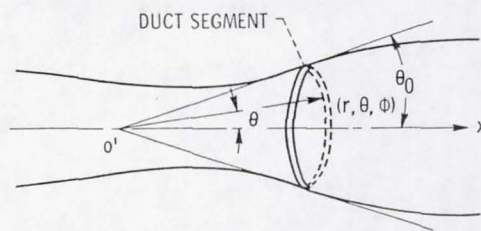


Figure 1. - Duct segment and local spherical coordinate system.

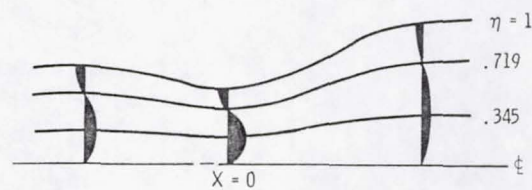


Figure 2. - Coordinate surfaces ($\eta = 0.345, 0.719, 1$) and transverse shape of (1, 1) mode at various axial locations.

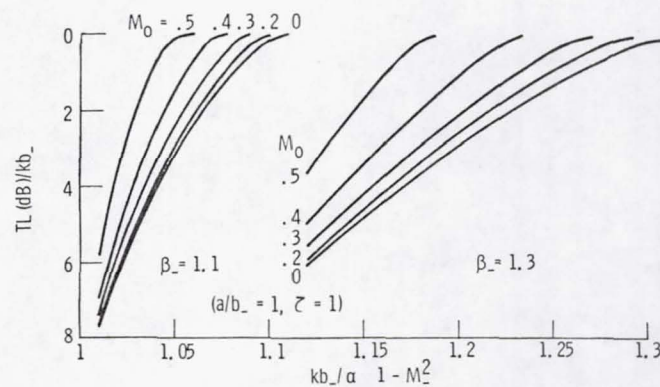


Figure 3. - Power transmission loss (TL) of (8, 5) mode, divided by kb_- , versus inlet cutoff ratio (γ_-).

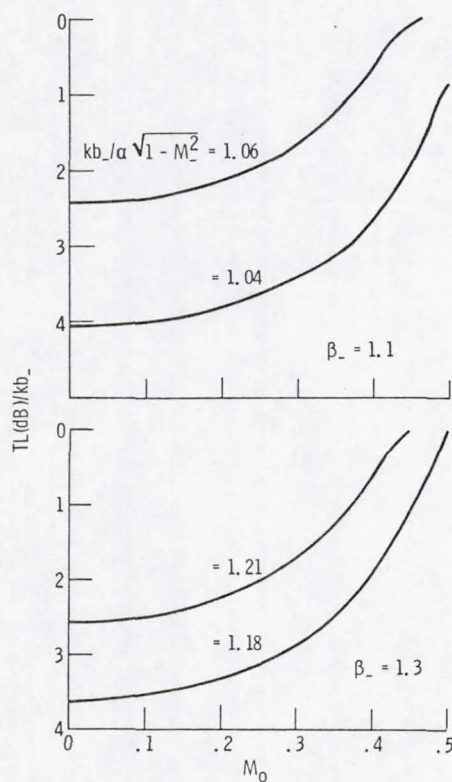


Figure 4. - Power transmission loss (TL) of (8.5) mode, divided by kb_- , versus Mach number at throat (M_0).

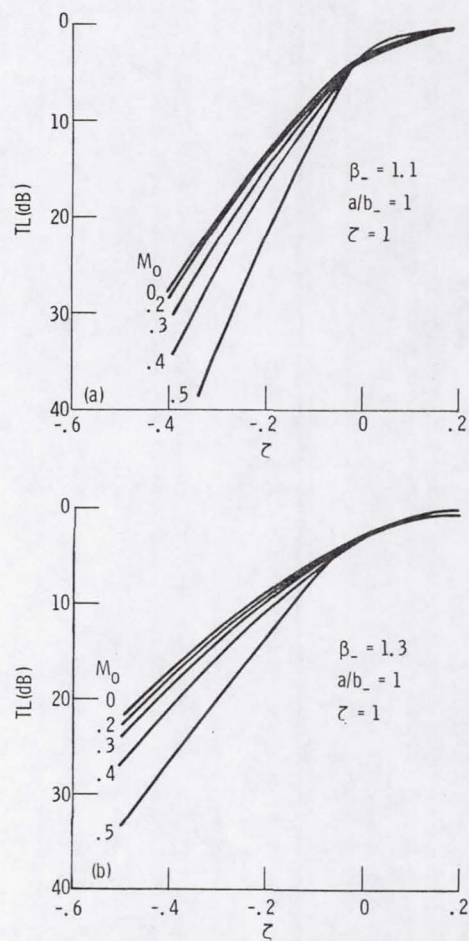


Figure 5. - Power transmission loss (TL) versus ζ . Each curve representing modes with eigenvalue greater than 3.

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16. Abstract A previously reported closed form solution is expanded to determine effects of isentropic mean flow on mode propagation in a slowly converging-diverging duct - a circular cosh duct. On the assumption of uniform steady fluid density, the mean flow increases the power transmission coefficient. The increase is directly related to the increase of the cutoff ratio at the duct throat. With the negligible transverse gradients of the steady fluid variables, the conversion from one mode to another is negligible, and the power transmission coefficient remains unchanged with the mean flow direction reversed. With a proper choice of frequency parameter, many different modes can be made subject to a single value of the power transmission loss. The paper also describes a systematic method to include the effects of the gradients of the steady fluid variables.			
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