NASA Reference Publication 1125

November 1984

Space Shuttle Wind Tunnel Testing Program Summary

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(NASA-RP-1125) SPACE SHUTTLE WIND TUNNEL TESTING PROGRAM (NASA) 492 p HC A21/MF A01 CSCL 22B
N85-11127
Unclas
H1/18 20562

NASA

NASA
Reference
Publication
1125

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National Aeronautics
and Space Administration

Scientific and Technical
Information Branch

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NOMENCLATURE

Symbols and abbreviations

$\bar{c}$	Mean aerodynamic chord, MAC
CF_4	Freon
g	Acceleration due to gravity
h	Altitude
He	Helium
K	Thousand
L_B	Body length (of Orbiter)
m	Million
M	Mach number
MAC	Mean aerodynamic chord, $\bar{c}$
N_2	Nitrogen
q	Dynamic pressure
Re	Reynolds number, based on Orbiter body length
sec	Seconds
V	Velocity
$\bar{V}'$	Viscous parameter
X_o	Body axis, longitudinal (Orbiter)
Y_o	Body axis, lateral (Orbiter)
Z_o	Body axis, vertical (Orbiter)
α	Angle of attack
β	Angle of sideslip
ϕ	Bank angle
γ	Flight-path angle
Γ	Universal gas constant
Λ	Sweepback angle

Abbreviations, Acronyms:

A	Aerodynamics - test type designator, or Ames Research Center - test responsibility designator
AADS	Ascent air data system
ABPS	Air breathing propulsion system
ADDB	Aerodynamic Design Data Book
ADS	Air data system (Orbiter)
AEDC	Arnold Engineering Development Center
ALT	Approach and landing test
AOA	Abort to once around
ASRM	Abort solid rocket motors
ATP	Authority to proceed (management milestone)
C	Carrier - test configuration designator
CAL	Cornell Aeronautical Laboratory
CDR	Critical design review (management milestone)
CG	Center of gravity
CR	Contractor report

DATAMAN	Data Management System (Chrysler Corporation)
DCR	Design certification review (management milestone)
DFI	Development flight instrumentation
ET	External tank
EPS	Electrical power subsystem
F	Marshall Space Flight Center - test responsibility designator
FCF	First captive flight (ferry program)
FMCF	First manned captive flight (ferry program)
FMOF	First manned orbital flight
FRL	Fuselage reference line
FRSI	Felt reusable surface insulation (blanket)
F.S.	Fuselage station, or Full Scale
H	Heating - test type designator
HRSI	High (temperature) reusable surface insulation (tiles)
I	Integrated Vehicle - test configuration designator
IV	Integrated Vehicle
L	Langley Research Center - test responsibility designator
L.E.	Leading edge
LRSI	Low (temperature) reusable surface insulation (tiles)
LTV	Ling-Temco-Vought Corporation
M	Johnson Space Center - test responsibility designator
MCR	Modification change request
MECO	Main engine cut off
NASA	National Aeronautics and Space Administration
-ARC	Ames Research Center
-FRC	Flight Research Center
-HQ	Headquarters
-JSC	Johnson Space Center
-LaRC	Langley Research Center
-LeRC	Lewis Research Center
-MSFC	Marshall Space Flight Center
NSWC	Naval Surface Weapons Center
O	Orbiter - test configuration designation
O/ET	Orbiter/external tank mated configuration
OFT	Orbital flight test
OML	Outer moldline
OMS	Orbital maneuvering system
OV	Orbiter Vehicle
PDR	Preliminary design review (managment milestone)
PRR	Preliminary requirements review (management milestone)
RCS	Reaction control system
RI	Rockwell International
RTLS	Return to launch site (abort)
S	Solid rocket booster - test configuration designator, or Structural dynamics - test type designator
SAL	(Space) Shuttle approach and landing
S&C	Stability and control
SCA	(Space) Shuttle carrier aircraft
SEP	Separation
SOF	(Space) Shuttle orbital flight
SRB	Solid rocket booster
SRM	Solid rocket motor
SRR	(Space) Shuttle requirements review (managment milestone)
SSECP-WTP	Space Shuttle Engineering Coordination Panel - Wind Tunnel Panel

SSME	Space Shuttle main engines
SSPO	Space Shuttle Project Office
SSV	Space Shuttle Vehicle
STAR	Scientific and Technical Aerospace Reports
STS	Space Transportation System
T	External tank - test configuration designator
TAMU	Texas A&M University
TBC	The Boeing Company
T.E.	Trailing edge
TPS	Thermal protection system
TVC	Thrust vector control
V/STOL	Vertical/short take off and landing
WL	Waterline
WTO	Rockwell International - Wind Tunnel Operations Group
Δ	Increment (management milestone)

ABSTRACT

A major phase of the Space Shuttle Vehicle (SSV) Development Program has been the acquisition of data through the Space Shuttle Wind Tunnel Testing Program. During the early concept studies it became obvious that the large number of configuration/environment combinations would necessitate an extremely large wind tunnel testing program. To make the most efficient use of available test facilities, and to assist the prime contractor for Orbiter design and Space Shuttle Vehicle integration, a unique management plan was devised for the design and development phase (1972 through 1983).

A brief overview of the Space Shuttle Program is given together with the evolutional development of the Shuttle configuration. A detailed review is provided of the wind tunnel testing rationale and the associated test program management plan and its overall results. Also given is information as to the various facilities and models used within this program. A unique posttest documentation procedure is presented. In conclusion, a summary of the types of tests per disciplines, per facility, and per model are presented together with detailed listing of the posttest documentation which is available through Scientific and Technical Aerospace Report (STAR).

PREFACE

Conceptual studies for a United States Space Transportation System (STS) determined that the large number of configuration/environment combinations would necessitate an extremely large wind tunnel testing program. To efficiently use the available test facilities and to assist the prime contractor for Orbiter design and Space Shuttle Vehicle integration, a unique management plan was devised for the design and development phase (1972 through 1983). This paper presents a detailed review of the wind tunnel testing rationale and the associated test program management plan. Included are tables of the complete wind tunnel program indicating facilities used. Also included are tables of the models, and the overall configuration schedules. The management plan involved facility representatives from the major testing centers. They were briefed regularly on the configuration status and the corresponding test requirements. In addition, the need for an integrated system for processing large blocks of wind tunnel data and for a standardized documentation concept was recognized and incorporated, utilizing automated data handling. Both the management system and the documentation methods resulted in reliable data and an efficient wind tunnel program for the Space Shuttle Vehicle.

This document should be a guide for future conceptual test planning of wind tunnel programs similar to that for the Space Shuttle. Items addressed are wind tunnel test planning and management, data management and documentation, new supplemental testing techniques, and innovations in model design.

1. INTRODUCTION

In late 1960, the National Aeronautics and Space Administration (NASA) explored the feasibility of employing a reusable vehicle for access to low-Earth orbit. Ideally, the desired vehicle would be similar to a conventional aircraft in both design and operation. This would minimize the expense of a single-use launch vehicle. To assess the vehicles' aerothermodynamic performance, concept feasibility studies in Phase A were conducted. The empirical data was relied on for generalized trade-off studies of the configurations selected. Phase B (definition) would rely on simple scaled models by each competing contractor. These scaled models were tested in the more critical aerothermodynamic areas. They were also supplemented by parallel NASA studies on generic configuration trade-offs. Phase C/D would then take the selected contractor, with an essentially frozen configuration, through the design and development stages.

The early feasibility studies determined that many complex configurations would be required to meet the objectives of a reusable vehicle. These configurations would entail an enormous amount of wind tunnel tests. Regardless of which configuration/operational concept combination was selected, many unique configurations from the viewpoint of ground facility tests and model requirements would result. The launch vehicle, whether completely reusable or only partially reusable, would change configuration with each staging. The entry vehicle would have large center of gravity shifts because of the varied payloads to be carried. It also would be subjected to the largest range of velocity, configuration, and environmental combinations ever experienced by a manned vehicle. In addition, requirements for horizontal low-speed flight tests and a "ferry" concept added several new configurations. All of these many combined shapes would require testing for aerodynamics, aerothermodynamics, structural loads, structural dynamics, and stage separation. Thus, even before the STS concept or the prime contractor was chosen, estimates for the amount of wind tunnel facility testing, exceeded that of any other previous aircraft or space vehicle program. To meet these many varied requirements, a new approach was required to coordinate the large number of tests. Also required was efficient use of the available test facilities and developing economical and timely reporting concepts. This would result in reliable data for the design analysts.

This paper (an extension of reference 1) presents the rationale for configuration testing within the basic disciplines. It also describes the wind tunnel program management approach and the documentation procedure. Included are detailed tabular listing of all of the testing that was performed to define the baseline configurations. Also included is the direct-support testing done by the NASA centers. It should be noted that the scope of the material in this paper is intended to cover the prime contractor SSV wind tunnel test program only. That is, the Phase C/D portion which extended from mid-1972 to late 1983.

2. BACKGROUND

The preliminary investigations of STS concepts, entitled Integral Launch and Recovery Vehicle Studies, commenced in February 1969, by direction of NASA's Office of Manned Space Flight. Within these studies the major aircraft manufacturers were invited to submit their concepts for evaluation. From these Phase A feasibility efforts, it was concluded that the lifting body, or so called "wingless" class vehicle, would not be compatible with efficient cargo packing and the necessary subsystem arrangements. Nor could this configuration provide the subsonic performance requirements that were needed, primarily lift-to-drag ratio. However, the proposed winged, two-stage vehicle concept appeared somewhat more promising. It satisfied the overall projected mission requirements. The major difficulty of this concept was the design and development of optimum aerodynamic configurations for the individual Orbiter and booster vehicles. Also required was a configuration for the integrated vehicle system. At this point in the program, there were two Space Shuttle Orbiter configuration concepts being evaluated. One was a straight wing design with a horizontal tail (ref. 2). The other was a delta wing design.

Subsequent to Phase A, funding for a follow-on preliminary design study (Phase B) was issued in July 1970 to the participating contractors. At this time the U.S. Air Force added their particular requirements. The major of these involved the areas of payload capability and crossrange requirements. The primary purpose of Phase B was for the contractors to further refine their proposed configurations including the new mission requirements. They also had to prepare a preliminary estimate of the costs. Because of the increasing complexity and expense of the Space Shuttle Orbiter/booster design studies, the contractors organized into teams of two. One concentrated on the Orbiter Vehicle; the other on the booster system. Shortly after, NASA management realized the large ultimate cost of the completely reusable concept. It decided to indefinitely delay the "flyback" booster in favor of an expendable booster. Additionally, it was decided to reduce the Orbiter size to be compatible with the chosen booster system. These design philosophy changes extended the Phase B effort for an additional year. This contract period was referred to as "Phase B Prime."

Following the Phase B Prime study, another extension was initiated. It reduced the fully reusable Space Shuttle Orbiter size further and concentrated on a two-stage, parallel-burn booster system concept. This booster concept was configured as a pair of recoverable 156-inch diameter solid rocket boosters (SRB), with an external liquid fuel tank (ET). They would feed the rocket engines in the Orbiter. This second extension was referred to as "Phase B Double Prime." The configuration results of this phase are shown in Figure 2.1.

In March 1972, NASA issued a request for formal proposals for the design and development of the Orbiter including systems integration, the ET system, and the SRB system. The Rockwell Rocketdyne Division had already been chosen to develop and produce the Space Shuttle main engines (SSME's), in July 1971. In July 1972, Rockwell International (formerly North American/Rockwell) was selected as the prime contractor for the Phase C/D design, development and production of the Orbiter, and the overall integration of the SSV system. In August 1973, Martin Marietta Corporation was awarded the contract for design, development, test, and production of the liquid-fueled ET. In November 1973, Thiokol Chemical Corporation was awarded the contract for the SRB's.

The carrier vehicle concept for ferry and air launch (low-speed flight tests) was originated in late 1973. The original concept of "bolt-on" air-breathing engines had operational limitations with range, increased turnaround time, and recovery from contingency bases. Technical concerns were scar weight to the Orbiter, thermal protection system (TPS) degradation, and possible cargo bay contamination. Studies done by the NASA-Flight Research Center (FRC) as well as other independent studies showed the carrier concept was feasible for both ferry purposes and the approach and landing test (ALT) program. In June 1974, the Boeing 747 was chosen to be the Shuttle carrier aircraft (SCA).

A historical account of the early programs leading up to the SSV concept can be found in reference 3.

3. CONFIGURATION EVOLUTION

Identification of the SSV configurations as they evolved can be confusing because of several levels of configuration designation. The initial designations, related to the program milestones, used the following acronyms:

ATP - Authority to Proceed
PRR - Preliminary Requirements Review
PDR - Preliminary Design Review
CDR - Critical Design Review

On the design engineering level, the prime contractor configuration control drawing designations were used. At times the lines (contour) drawing designations were all that was available because changes were made so rapidly. In addition, SSV designations 1 through 6 were used by both management and engineering. Figure 3.1 shows the approximate time periods for the various designations. Figure 3.2(a) and 3.2(b) are summaries of the major configuration definitions for the Orbiter and Integrated Vehicle (IV), respectively. Figure 3.3(a) through 3.3(e) are three-view drawings of the major Orbiter definitions shown in fig. 3.2(a). Figure 3.4(a) through 3.4(e) are three-view drawings of the major IV definitions shown in fig. 3.2(b). Figure 3.5(a) and 3.5(b) are three-view drawings of the OV-102 and IV for the STS-1. Tables 3.1 through 3.3 give the dimensional parameters for OV-102, the ET and the SRB's, respectively. Table 3.4 presents the associated reference areas and lengths used to normalize the aerodynamic forces and moments.

The ATP Orbiter aerodynamic shape had a 50° sweep delta wing planform sized to provide 150 knots (77.2 m/sec) design touchdown speed with a 40,000-pound (18,100 kilogram) return payload. Elevons were sized to provide trim at hypersonic speeds over an angle of attack range from 20° to 50° with an operational center of gravity (CG) range of 3 percent body length (LB). The cargo bay was 15 feet in diameter (4.57 meters) by 60 feet long (18.2 meters) to accept a wide variety of payloads. The remote manipulator arms were stowed in a dorsal fairing along the top of the payload bay doors. For ferry and entry assist, an air breathing propulsion system (ABPS) was situated in the aft portion of the payload bay. Three main propulsion system rocket engines were located at the base of the aft fuselage and orbital maneuvering systems (OMS) engines were installed in two removable pod modules on the side of the aft fuselage. Reaction control system (RCS) rocket engines were also located in the aft pods and on the forward fuselage. The pilot's eye (cockpit location) was 208 inches (528.3 cm) aft of the nose and had an angular view of 20° up and 24.5° down. The nose radius was 25 inches (63.5 cm) and blended smoothly into the low fineness ratio body. The ATP Integrated Vehicle had the Orbiter attached to the ET "piggy-back" style with the Orbiter nose 80.3 feet (24.48 m) aft of the ET nose. The Orbiter fuselage reference line (FRL) is canted down so that the Orbiter is oriented at a -1.2° incidence with respect to the ET centerline. The SRB's are attached to the ET such that the noses of each SRB are 17.5 feet (5.33 m) aft of the ET nose and 3.1 feet (0.94 m) above the ET centerline. The centerlines of the SRB's and the ET are parallel. Two ASRM's (abort SRM's) are mounted at the aft end of the Orbiter body. The ET is essentially a cone-cylinder arrangement. It is fitted with a retro SRM package at the tank nose to facilitate ET deorbit. The external shape of the retro SRM is a small hemisphere-cylinder with a nose radius of 20.5 inches (52.07 cm). It has a length of 124 inches or 10.33 feet (3.15 m). The conical nose portion of the tank has a semivortex angle of 30° which blends smoothly into the cylindrical section of the ET. The shoulder blending radius at the cone-cylinder juncture is the same as the cylinder radius, 159 inches (403.9 cm). The ET overall length is 182.0 feet (55.47 m). The nose radius of the SRB's is 13 inches (33.0 cm) and the cone semivortex and angle is 18°. The cylinder diameter is 156 inches (3967.2 cm), and the overall SRB length is 184.8 feet (56.33 m). The fixed nozzles are canted outward 11° in the yaw plane so that the boosters will be thrusting through the approximate center of gravity of the vehicle during the boost phase.

The PRR configuration evolved from the ATP configuration based on MCR 0026 (Master Change Record) in October 1972. The most obvious changes follow. The OMS pods were rotated from the aft fuselage side to the aft body shoulder and lengthened slightly. The canopy was moved aft approximately 52 inches (132.1 cm) with an angular view of 7° up and 18° down. The forebody was redesigned to accommodate internal packaging revisions and to improve the transition to the midbody. Wing refinements included an increased thickness ratio, a slight leading-edge droop and minor wing body fillet modifications. The Orbiter incidence was increased to +0.5° to improve the ET separation performance and the IV trim angle. The abort SRM's were deleted. The air breathing propulsion for landing assist following orbital flights was deleted. The ET nose was changed to an ogive shape to reduce the drag. The SRB's were shortened and moved aft resulting in a slightly longer integrated vehicle. Most of the positioning of the elements (Orbiter and SRB's) relative to the ET, were to improve the element-to-element interference drag. It also alleviated the SRB plume effects on the Orbiter base. The SRB's also had an expanded shroud, had thrust vector control (TVC) added with a reduced precant, and had the aft strakes removed.

Vehicle 2A, also referred to as the "150K Orbiter," (where K indicates 1000 pounds) encompassed the largest changes of all (MCR 0074). Basically the vehicle dry weight and payload down weight were reduced significantly, requiring a complete resizing of the Orbiter. A 45°/79° double-delta wing planform was incorporated with reduced glove leading-edge radius and forward sweep to the trailing edge. It included wing twist, camber, and incidence revisions for improved subsonic performance. Improved low-speed performance and a reduced static margin requirement permitted a reduction in wing size to 2,690 square feet (250 square meters) and resulted in rebalancing of the OV to meet stability and control requirements. Nose camber and radius, body cross section, and upward sloping forebody slab sides were selected to improve hypersonic pitch trim and directional stability. By combining them with wing-body blending, entry heating was reduced on the body sides. These changes also simplified the nose structure. The CG travel requirement was reduced from 3 to 2 percent body length. The ferry air breathing engines were moved to a position under the wing (not shown). The SRB's were shortened and moved aft further resulting in a shorter overall vehicle. The SRB yaw gimbal setting was reduced to 0° and the nozzle flare angle was reduced. The ET was also shortened and the Orbiter was repositioned on the ET near its original location.

Vehicles 3 and 4 were essentially the same from an aerodynamic configuration viewpoint. Vehicle 3 was initiated with MCR 0200, with seven revisions carrying through the start of Vehicle 4. Early changes shortened the body by 38 inches or 3.17 feet (0.97 m) and smoothed the body nose area while incorporating a smaller nose radius at the same time. The wing glove leading edge was increased to 81° and the incidence was decreased from 3° to +1/2°. Also some minor airfoil changes were made, the wing was lowered 4 inches (10.2 cm), and the lower body was refaired. In addition the bodyflap span was reduced. The primary purpose of these changes was to improve the overall aerodynamic and aerotheromdynamic performance. The CG range requirement increased to 2.5 percent body length to allow 1/2° for aerodynamic trim uncertainties with a new payload down of 32,000 lbs (14,515 kg). In addition, the manipulator arm dorsal fairing along the top of the payload bay doors was deleted. The manipulator was stowed inside the payload bay. Later Orbiter changes (mid-1973) included thickening the airfoil 6 inches (15.24 cm) at the elevon hingeline. In early 1974 the ferry air breathing engines were deleted. The SRB's were moved forward relative to the ET as was the Orbiter. The FT was shortened and the retro package (spike) removed. The result was a slightly shorter overall vehicle length.

The Vehicle 5 Orbiter had the OMS pods shortened and refaired to clear the payload bay doors in early 1974 (MCR 0500). Other lesser external shape changes modified the wing tips, increased the elevon gaps, and deleted the vertical tail drogue chutes. Later (mid-1975) changes added recessed thermal glass in the windshield, observation windows, and hatch windows. Also the covers were removed from the forward RCS ports and from the umbilical doors in the aft body. The ET and SRB length changes as well as the distance of the SRB and Orbiter aft of the ET, were in approximately 4 feet or less. An ascent air data system (AADS) in the shape of a cone was added to the tip of the ET.

Vehicle 6 had no significant external shape differences from Vehicle 5.

A thorough description of the design logic to optimize the Orbiter aerodynamic configuration can be found in references 4 and 5. Reference 6 discusses the role of the entry aerothermodynamic environment on the Orbiter design, and reference 7 presents some of the challenges to the structural dynamicist.

TABLE 3.1 - ORBITER DIMENSIONAL PARAMETERS

Component	Parameter	Value
Total vehicle body + exposed wing)	Reference area, ft^2 (m^2)	2690 (249.91)
	Planform area, ft^2 (m^2)	3952 (367.14)
	Surface wetted area, ft^2 (m^2)	11136 (1034.53)
	Length overall, ft (m)	122.0 (37.18)
	Reference length, ft (m)	107.5 (32.76)
	Height, gear up, ft (m)	46.14 (13.75)
	Height, on gear (static), ft (m)	53.76 (16.38)
	Span, ft (m)	78.056 (23.79)
Body	Reference length (nose at X_o=238 inches), in. (cm)	1290.3 (3277.36)
	Depth, maximum (X_o=1280 inches) ft (m)	19.92 (6.07)
	Width, maximum (X_o=1528.3 inches) ft (m)	22.0 (6.70)
	Planform area, ft^2 (m^2)	1914.4 (177.85)
	Surface wetted area, ft^2 (m^2)	5634 (523.39)
	Base area (includes OMS pods), ft^2 (m^2)	436.7 (40.57)
	Cargo bay (diameter by length), ft (m)	15 by 60 (4.57 by 18.28)
Wing (includes body carry through)	Planform area, ft^2 (m^2)	2690 (249.91)
	Span, ft (m)	78.056 (23.79)
	Aspect ratio	2.265
	Taper ratio	0.20
	Sweep, leading edge, deg	45
	Sweep, trailing edge, deg	-10
	Dihedral (at wing trailing edge), deg	3.5
	Root chord (Y_o=0 inches)	
	(theoretical) Length, ft (m)	57.44 (17.50)
	Quarter chord station, X_o, inches (cm)	1008.31 (2561.10)
	Tip chord (Y_o=468.34 inches)	
	Length ft	11.48 (3.50)
	Quarter chord station, X_o, inches (cm)	1338.80 (3400.55)
	Incidence, deg	+0.5
	Airfoil section	0012-64 modified
	MAC, $\bar{c}$ (Y_o=182.13 inches)	
	Length, ft (m)	39.56 (12.05)
	Quarter chord station, X_o, inches (cm)	1136.83 (2887.54)

TABLE 3.1 - CONTINUED

Component	Parameter	Value
Wing, exposed	Planform area (including glove), ft^2 (m^2)	2012.4 (186.95)
	Surface wetted area, ft^2 (m^2)	4001.2 (371.71)
	Root chord (Y_o=108.0 inches)	
	Length (inc. glove), ft (m)	80.83 (24.63)
	Quarter chord station, X_o, inches (cm)	778.5
	Incidence, (Y_o=199.045 inches), deg	-0.5
	Airfoil section (Y_o=199.045 inches)	0.0010 modified
	Leading edge glove, deg	81
Elevon (one side)	Area, ft^2 (m^2)	206.57 (19.18)
	Span, ft	28.87 (8.79)
	Aspect ratio	4.03
	MAC, $\bar{c}$ length, ft (m)	7.46 (2.27)
	Quarter chord station, X_o, inches (cm)	1409.375 (3579.81)
	Distance from elevon centroid to hinge line, inches (cm)	44.75 (113.665)
	Deflection (elevator/aileron), deg	+20, -35
	Inboard/outboard split line, (Y_o = 311 inches)	
Vertical tail (includes rudder and speed brake)	Planform area, ft^2 (m^2)	413.25 (38.39)
	Span, ft (m)	26.31 (8.02)
	Aspect ratio	1.675
	Taper ratio	0.404
	Sweep, leading edge, deg	45
	MAC, $\bar{c}$ length, ft (m)	16.65 (5.07)
	Elevation, Z_o, inches (cm)	635.5 (1614.17)
	Quarter chord station, X_o, inches (cm)	1463.35 (3716.90)
	Root chord length, ft (m)	22.37 (6.81)
	Tip chord length, ft (m)	9.04 (2.75)
	Airfoil section (root - tip)	10° sym 60%-40% double wedge
	Sweep, trailing edge, deg	26.2

TABLE 3.1 - CONCLUDED

Component	Parameter	Value
Rudder and speedbrake	Planform area, ft^2 (m^2)	97.84 (9.08)
	Span, ft (m)	16.55 (5.04)
	$\bar{c}$ length, ft (m)	6.07 (1.85)
	Elevation, Z_o, inches (cm)	670.41 (1702.84)
	Quarter chord station, X_o, inches (cm)	1575.77 (4002.45)
	Deflection, rudder (maximum), deg	22.8
	Deflection, speedbrake, deg	0 to 87.2
	Hinge Line, deg	34.83
Body Flap	Planform area, ft^2, (m^2)	135.75 (12.61)
	Fuselage station of hinge line, inches (cm)	1532 (3891.28)
	Span (equivalent), inches (cm)	241.33 (612.97)
	Chord, inches (cm)	81 (205.74)

TABLE 3.2 - ET DIMENSIONAL PARAMETERS.

Parameter	Value
Length overall (OML), ft (m)	154.4 (47.05)
Ogive radius (OML), inches (cm)	613.0 (1557.02)
Cylinder diameter (OML), inches (cm)	333.0 (845.82)
Base area (OML), ft^2 (m^2)	604.8 (56.18)

TABLE 3.3 - SRB DIMENSIONAL PARAMETERS

Parameter	Value
Length overall, ft.(m)	149.1 (45.44)
Nose radius, inches (cm)	13.28 (33.73)
Cone semi-vertex angle, deg	18.0
Cylinder outside diameter, inches (cm)	146.0 (370.84)
Base area, ft^2 (m^2)	235.0 (21.83)
Nozzle deflection angle:	
Null position, pitch & yaw, deg	0
Gimbal range, pitch & yaw	
TVC axes, deg	± 5.0
Body axes, deg	±8.0

TABLE 3.4 - AERODYNAMIC FORCE AND MOMENT REFERENCE DIMENSIONS.

Parameter	Reference Value
Longitudinal and lateral/directional coefficients	
Wing Area, ft^2 (m^2)	2690.000 (249.91)
Wing Span, ft (m)	78.057 (23.79)
MAC, $\bar{c}$ length, ft (m)	39.568 (12.06)
Hinge moment coefficients	
Elevon	
Area, S_e, ft^2 (m^2)	210.000 (19.51)
Chord $\bar{c}_e$, ft (m)	7.55 (2.30)
Bodyflap	
Area, S_{BF}, ft^2 (m^2)	135.000 (12.54)
Chord, $\bar{c}_{BF}$, ft (m)	6.75 (2.05)
Rudder/speedbrake	
Area, S_r, ft^2(m^2)	100.150 (9.30)
Chord, $\bar{c}_r$, ft (m)	6.10 (1.86)

Note: The aerodynamic moments for the Orbiter were reduced at a point 65 percent aft of the Orbiter nose and on the Orbiter centerline. For the Integrated Vehicle the moments are reduced at a point coincident with the Orbiter nose and on the external tank centerline.

4. TEST CONDITIONS AND REQUIREMENTS

The nominal mission phases dictate distinct flight conditions to which the SSV is subjected. These phases can be divided into the three major operational areas: launch, separation, and entry. The flight environment of the various configuration combinations is dependent on the vehicle attitude, velocity, and the accompanying flowfield interaction with the configuration. A sketch of the nominal mission phases is shown in fig. 4.1. Trajectory characteristics for a typical ascent are shown in fig. 4.2(a) and for a typical entry in fig. 4.2(b).

In the launch phase the vehicle is positioned vertically on the launch pad (fig. 4.3). Before and during lift-off, the effect of ground winds must be considered. The main engine noise (vibration loads) and pad overpressures experienced at ignition can also be important. As the vehicle ascends, the aerodynamic stability and control characteristics are important. However, they are not critical because of the overpowering thrust of the vectorable main engines and SRB's. At an early point however, the aerodynamic loads become large (in the transonic flight regime) with the additional possibility of empennage buffeting. As the launch speed increases, aerodynamic heating, and, in particular, localized heating between the vehicle's individual elements also increases. Throughout launch, the total vehicle drag including the effects of the engine plumes must be determined. Some of the more important IV aerodynamic considerations are noted in fig. 4.4.

For nominal flight, the two distinct element separation phases are SRB separation near Mach 5.0 and ET separation near orbital insertion. ET separation problems are actually more significant aerodynamically for the abort phase referred to as "return to launch site" (RTLS). Each separation phase must consider the aerodynamic interference, or proximity, effects to avoid element recontact. The aerodynamics, airloads, and aeroheating of the individual SRB and ET elements are required for ET disposal studies and SRB recovery analyses. In addition, SRB recovery involves determination of the characteristics of a suitable parachute system. Fig. 4.5 depicts the separation characteristics. Fig. 4.6 shows the events for SRB recovery.

During the entry phase, the Orbiter will descend from a Mach number near 28 to a landing speed near 200 knots (102.9 m/sec). At the upper level of the Mach regime, aeroheating, and stability and control considerations also involve the RCS interactions. Throughout entry the effectiveness of all of the Orbiter control surfaces (elevons, bodyflap, rudder, and speedbrake) must be determined. This is because the attitude profile changes from a high angle of attack at high Mach numbers (to dissipate the heat and slow the vehicle) to a conventional "airplane-like" attitude for landing. The subsonic lift-to-drag ratio must be large enough so that the Orbiter speed and angle of attack are not excessive at landing. In addition, ground effects during landing are important. An additional characteristic of the Orbiter during entry is the large range of CG locations (because of payload placement) that the vehicle's stability and control characteristics must contend with. Some of the more important Orbiter aerodynamic considerations are shown in fig. 4.7.

For the carrier program, several distinct modes of operation were required. First, in the ferry mode, the Orbiter/SCA in the mated configuration would perform the ferry mission up to a range of 2000 nautical miles (3706.2 km).

Other potential ferry missions would involve the ET or a payload cannister. For the airlaunch, or the ALT mission, the mated configuration would climb to altitude (approximately 30,000 ft. or 9144 m) and launch the Orbiter in a "top-launch" concept. Initial ALT flights would be performed with a tailcone or fairing that covered the Orbiter main engine nozzles and stayed attached to the Orbiter during the free-flight portion. Later ALT flights would be performed without the tailcone. The most complex flight condition to determine for ferry was the definition of the wake caused by the blunt-based Orbiter and the subsequent effect on the 747 vertical tail. For the ALT mission the separation procedures and characteristics were the most difficult to determine. Fig. 4.8 shows the Orbiter/SCA mated configuration with the Orbiter to SCA incidence to be set for either a ferry or launch mission. Fig. 4.9 shows the ALT Orbiter Vehicle (OV101) and some of its special characteristics for the ALT program.

5. WIND TUNNEL TESTING RATIONALE

Much of the expertise in determining what type of wind tunnel testing that would be required was developed during the early SSV studies (see refs. 8 through 10). In addition, the selected contractor, Rockwell International, had a large amount of experience with previous aircraft test programs. It also had experience with the Apollo test program. Several differences for the SSV program were apparent from the beginning. However, these differences dictated that the SSV wind tunnel program be very thorough and at the same time be highly efficient. Thoroughness was dictated by the fact that STS-1 would be orbital and would carry a crew. There were no plans for using the conventional graduated flight test approach that new aircraft normally use. New ground was being broken in all flight phases and little empirical data were available for the early SSV studies. Efficiency was dictated by both the design and development schedule and by the need to keep costs to a minimum. Each distinct configuration/environment test requirement in each discipline had to be justified. Testing had to be designed to obtain reliable data. To minimize program costs, the major portion of the wind tunnel program would be conducted in NASA facilities. The basic objective of the overall wind tunnel program was to conduct the mainstream testing that would meet the technical requirements of the SSV aerodynamic, aerothermodynamic, airloads, structural dynamics, and separation disciplines. Adhering to this procedure would result in a safe and successful operational SSV. In general, the early testing was done to define basic vehicle characteristics and parametric effects. The latter data was used incrementally to estimate the characteristics of any proposed design modifications. When the configuration definition was "frozen," for management review purposes, then the new design was tested to verify the estimated data base. This procedure was repeated several times throughout the program, while holding any extensive verification tests to a minimum until the end of the program when the "as built" configurations could be tested. The testing rationale for each of the aforementioned test disciplines is discussed in the following paragraphs as applicable to the major configuration/operational flight phase combinations for the Integrated Vehicle, Orbiter, and ferry configurations.

Integrated Vehicle

The basic aerodynamic force and moment data were needed early. This determined the requirements for both SRB and SSME engine-on additions to the total vehicle stability. Total vehicle stability required that the aerodynamic and thrust, forces and moments be in equilibrium. The aerodynamics would dictate SRB and SSME precant nozzle angle settings and engine gimbal requirements. In addition, because the plume effects have such a strong effect on stability, the early tests included plume simulations. However, in testing with such a complex launch configuration base area there would obviously be sting interference effects. To account for these, engine-off tests were performed. A conventional sting-out of the aft end of the ET model measured force and moment data, elevon and rudder hinge moments, attach structure interface loads, wing moments (bending, torsion, and shear), and effects forward of the base area (fig. 5.1). Similar measurements were made with the Orbiter as the sting-supported element (fig. 5.2) allowing for sting effects to be accounted. Later tests, specifically for incremental model support interference effects, were performed to address all of the sting arrangements used (fig. 5.3). For plume effects, early tests were run with the conventional aft-sting arrangement (fig. 5.4). Later tests were run using a blade strut, from the lower surface of the ET, to concentrate on measurements in the base region (fig. 5.5). The blade mount provided a relatively "clean" base region for high quality measurement of base effects. Early exploratory testing used analytically-determined solid plume shapes (varying with Mach number, altitude and other key jet simulation parameters) as shown in fig. 5.6.

Compromises had to be made in the testing program when duplication of the actual engine exhaust plumes was planned. It was neither technically nor economically feasible to completely duplicate the exhaust gas from the SRB's and SSME's for each of the SSV launch vehicle tests for the following reasons.

a. The geometry could not be accurately simulated because of the necessary plumbing required to pass the simulant gases into the model.

b. Since the base area is the primary area affected by the launch vehicle plumes, a blade support system mounted through the ET is required. This is used rather than a sting support system to properly model the base area. The blade support system will invalidate aerodynamic data only at large aerodynamic angles.

c. Exhaust plume testing is an order of magnitude more expensive and time-consuming than "standard" aerodynamic testing.

The approach used in the SSV test program was to use state-of-the-art techniques for the basic power-off data base. Then generate power effect increments from the limited exhaust plume tests. The exhaust simulant gas used in these tests was high-pressure, unheated air. Using air as the simulant gas was acceptable based on the results of a plume technology study program. This separate program was designed to establish a set of simulation parameters. These parameters would correlate wind tunnel derived base pressures, using air as the simulant gas, with the expected results from the prototype vehicle. The resulting simulation parameter, a function of plume shape and gas dynamics characteristics, was applied to the scaled SSV exhaust plume test data. This was done to obtain the base and forebody plume effect increments. The tests covered the transonic and low supersonic region where the plume effects are most significant. Supplemental data from base heating tests were used to fill in the high Mach number data points. Such points as the vacuum chamber test arrangement as shown in fig. 5.7. In this way, many tests served several purposes and generated data in several separate test disciplines. Pressure tests (distributed loads tests) were accomplished in this manner throughout the program. These tests in addition to the extensive detailed testing to determine wing bending, torsion, and shear (as well as elevon-rudder hinge moments) were mostly done without the plume simulations because of the complications of having the instrumentation and the plumbing for the pressurized air all in the same model. The resulting distributed loads data were integrated to obtain forces and moments which were then compared to the test forces and moments. These two independent sets of data were compared and "balanced" to be within 3 percent of one another.

Static force and moment data on the SRB's and Orbiter/external tank (O/ET) configurations were obtained at Mach 4.5 for nominal staging conditions. The data was taken in two modes to reflect the flowfield environment encountered at staging. Namely, separation motors were simulated using high pressure air in conjunction with model nozzles scaled to reproduce jet-to-free-stream momentum ratio. Both SRB's were used with the O/ET model in plume-on testing to properly account for cumulative effects on the O/ET (fig. 5.8). Only a single SRB was used in the plume-off regime, the effects of the second SRB on the O/ET being derived analytically (fig. 5.9). Relative motion between the SRB's and the O/ET was produced by an automated captive trajectory system. This system was programed to sequentially vary the SRB relative positions according to a preprogramed run matrix. Artificial boundary-layer trips were not employed over the tested Reynolds number range of 1.4 to 7.0 x 10^6 per foot. This was because the presence of intervehicle shocks assured a turbulent boundary layer. A unique data organization strategy, the "hypercube" approach, was developed to facilitate use of the eight required independent variables (jet momentum ratio, O/ET pitch and yaw angles of attack, SRB relative longitudinal, vertical and lateral displacement, and SRB relative pitch and yaw orientation). The "hypercube" data strategy, as opposed to the classical grid data format, resulted in the reduction of required test data points by two orders of magnitude.

The nominal ET separation procedure is accomplished at an altitude at which aerodynamic forces and moments are negligible as compared to the forces and moments due to the RCS jets (used for the separation maneuver). Therefore no testing was done here. However, during the RTLS abort, the ET must be separated in a significant aerodynamic environment. The interaction of the RCS jets with the free-stream flow is substantial. The testing was accomplished for these flight conditions using the captive trajectory system much in the same manner as the SRB separation technique described above (fig. 5.10).

Postseparation aerodynamic forces and moments, as well as airload pressure distributions were obtained for the ET and for the SRB's at the appropriate test conditions. Both configurations required testing at high angles of attack because of tumbling after separation (ET for RTLS conditions). In addition the SRB recovery parachute system required some conventional testing. These tests were conducted using standard procedures.

Ascent aerothermodynamic heating tests began in the last half of 1973 after the configuration had gone through most of the major changes. The bulk of the testing was done using thermocouples in conjunction with thin-skinned models to measure rapid temperature changes (fig. 5.11). Pressure testing was done for the same test conditions to better define the local flow environment. Later testing concentrated in the base area identifying requirements for the base heat shield at high altitudes (fig. 5.12). These tests measured pressures and heat transfer characteristics with simulated plumes. Generally as time went on, the models had higher fidelity (such as that shown in fig. 5.13). The number of measurements increased concentrating in the critical areas. Supplementary testing with flat plate models duplicated areas that had configuration discontinuities, such as with the TPS tiles shown in fig. 5.14. Also, testing was done using oil-flow techniques to identify flow patterns and regions of high pressure concentrations (fig. 5.15).

Ascent structural dynamics testing centered around aeronoise (or fluctuating pressure) testing in the critical transonic and low supersonic regions of flight. Tests were also performed checking the possibility of flutter initiation, again during transonic/low supersonic flight, as shown in fig. 5.16. In addition, tests determining the effects of ground winds are shown in fig. 5.17.

In midprogram, subsystem managers of aerodynamics, airloads, heating, and stage separation felt that air data measurements would be required for the ascent phase for postflight analyses. These concerns resulted in an ascent air data system (AADS). Testing was done largely using a 7 percent forebody model (fig. 5.18) with supplementary tests on complete scale models for SSV element effects. Results of the AADS flight performance can be found in references 11 and 12.

Orbiter Configuration

The configuration for entry vehicle testing was not as complicated, as was the launch vehicle. In general, the same "balancing" procedure for force and moments, and distributed airloads that was used for the launch vehicle data was also used for the entry vehicle data. Some of the many model support arrangements are shown in fig. 5.19. A large scale model, shown in fig. 5.20, tested low speed characteristics with much of the surface discontinuities and outstanding features represented. Additional test areas for the entry configuration included.

a. Control surface deflections for the elevons, rudder, speedbrake, and bodyflap. Initial testing used "bolt-on" model parts that represented discrete deflection angles, figs. 5.21(a),(b). Later, more sophisticated models were made with internal mechanisms that allowed the control surfaces to be moved and set from outside the tunnel, fig. 5.21.(c). This procedure circumvented facility shut-downs for model changes and made occupancy hours much more productive (ref. 13).

b. RCS testing using simulated jet exhausts (as shown in fig. 5.22). Use of the RCS is critical during entry. Any adverse effects because of control surface pressure changes had to be identified and this information fed into the stability and control system. Some of the geometric moments produced by the jet engine forces were negated by induced jet effects on aerodynamic surfaces (wing and vertical tail). The effects of the various combinations of the main thrusters, positioned for pitch, yaw, roll moments, as well as for -Z translation (for ET separation) were tested. They concentrated on major aerodynamic degradation areas retesting where required.

c. Aeroelastic testing was used for effects of wing and vertical tail bending. Simple elastic wing and vertical tail model parts were attached to existing rigid models to determine the effects of the resulting shapes on the basic vehicle aerodynamics (fig. 5.23). These data were used to confirm the analytical predictions.

d. Ground effects were measured to ensure that control capability was available for landing. And more importantly they were measured to alert the crew to the expected aerodynamic effects during this critical flight phase. Tests were conducted initially with a fixed ground plane and later with a moving ground plane as shown in fig. 5.24. Low speed tests were also conducted to measure landing gear loads (fig. 5.25).

e. The Orbiter entry ADS was tested using 10 percent forebody models as shown in fig. 5.26. Test Mach numbers covered the flight operational range of the ADS, from Mach 0.20 to 3.5. Supplemental air data was taken during the large scale model tests, fig. 5.20(b). Details of the Orbiter ADS design, calibration and flight results can be found in reference 12.

Initial aerodynamic heating tests were conducted on the entry configuration with the emphasis on defining the overall environment. These tests were much more extensive than the integrated vehicle tests because of the harsher entry heating environment and because of the unique individual-tile TPS used on the Orbiter. The heat sensitive coating (phase change paint) technique was used to save time and money for the early configuration evaluation studies. Phase change coating models (a plastic-like substrate) are inexpensive and can be made more rapidly than the instrumented models (fig. 5.27). Heat transfer is determined by measuring the time required for a point on the surface of a model to reach the melting temperature of the thin coating. An added feature of this technique is that it is also a form of flow visualization. Another inexpensive technique that was used quite extensively was oil-flow photographs. They define streamline directions and local flow separation characteristics (fig. 5.28). Follow-on tests used the thermocouple/thin-skinned model technique or calorimeters to obtain more detailed data (fig. 5.29). Thermocouples were attached to the inner surface of the model at given points. Temperature time histories were taken with the angle of attack, Reynolds number, and Mach number as variables. Lower heating rates were measured with thin film gauges (resistance thermometer slug calorimeters) and the higher rates with thermocouple gauges (coaxial surface thermometers). Tunnel conditions were monitored with a dual probe that measured temperature and shock stagnation pressure (fig. 5.30). These thermocouple tests defined the temperature distributions around the Orbiter. Closely-related pressure tests defined pressure distributions for the same test conditions. Shadowgraph and Schlieren pictures of the flow patterns were useful in defining several necessary parameters, such as shock-standoff distance and boundary-layer flow conditions (fig. 5.31). Flat plate tests with full scale tiles determined the effect of tile gaps and surface irregularities as well as tile orientation (fig. 5.14). In addition, large scale testing was done in other critical areas such as leading-edge surfaces, elevon seal gaps, landing gear doors, and other areas where surface discontinuities would cause local "hot spots."

Structural dynamics testing was important. The validity of early configuration concepts was examined and critical areas defined, especially potential flutter tendencies of the Orbiter aerodynamic surfaces (fig. 5.32). The purpose of these tests was to acquire, early in the design process, experimental data in the transonic flight region to support analytical flutter predictions. Two models were used. One was scaled with the stiffness of the proposed baseline vehicle; a second with a reduced stiffness level. These results, with the aid of various computer programs, established flutter boundaries and substantiated proposed margins. All wing/elevon and fin/rudder models were designed to have variable control surface stiffness. This allowed exploration of potential coupling of the control surface with parent surface modes. These models also evaluated buffet and stall flutter tendencies. A semirigid flutter model was used to do the final evaluation of the Orbiter flutter boundary over the Mach/dynamic pressure range (fig. 5.33). Acoustic tests of the Orbiter surfaces covered by TPS had indicated that failure might be initiated by extreme pressure gradients. Gradients such as those produced by aerodynamic shocks, as well as structural vibrations resulting from acoustic or turbulent boundary-layer pressures. To evaluate the sensitivity of the TPS to these simultaneous effects, compression, and expansion shock tests were performed (fig. 5.34).

Ferry/ALT Configuration

Much of the feasibility testing for the ferry/ALT launch configurations was performed by the carrier vehicle contractors (Boeing and Lockheed) before the selection of the Boeing 747. Force and moment testing for detailed configuration development was still needed. Also needed was verification of the mated vehicle as well as the separation characteristics. In addition, testing was required to obtain a low drag tailcone for the Orbiter. This tailcone would minimize the buffet disturbance to the carrier aircraft. The force and moment tests were performed in the same facilities that the carrier aircraft contractor had used because of model compatibility and data comparability (fig. 5.35). Separation tests were done using a minimum matrix of conditions in conjunction with a computer graphics program. This program varied each vehicle's control settings (including spoilers, landing gear, etc.) to optimize on a safe separation procedure (fig. 5.36). Several exploratory tests were required to ensure the carrier aircraft vertical tail would be able to sustain any buffeting induced by the tailcone wake (fig. 5.37). In the process of defining an optimum tailcone configuration (low drag, low wake) many sting arrangements were utilized to minimize the model support effects (fig. 5.38).

6. WIND TUNNEL TEST PROGRAM MANAGEMENT

During the Phase A and B development programs, the NASA Johnson Space Center (JSC) and the NASA Marshall Space Flight Center (MSFC) monitored some four major contracts in their efforts to produce a suitable vehicle design to achieve the proposed space transportation mission. Near the end of Phase B, NASA Headquarters assigned JSC responsibility of managing the overall SSV integration task and the Orbiter development. MSFC had responsibility for the ET, the SRB's, and the SSME development. To effectively and efficiently manage the program, JSC established the Space Shuttle Program Office (SSPO) and created project offices representing the Orbiter, the ET, the SRB's, and the SSME. Each project office in turn assigned subsystem hardware and software managers. One of these was the Aerodynamics Subsystem Manager. In addition to responsibility for the development of the Space Shuttle aerodynamic data base, the Aerodynamic Subsystem Manager was also responsible for the organization and implementation of the overall SSV wind tunnel test program.

To assist in the execution of the Space Shuttle wind tunnel test program task, a committee of technical representatives was formed. It was referred to as the Space Shuttle Engineering Coördination Panel-Wind Tunnel Program (SSECP-WTP). Initial membership was requested from each of the NASA centers including NASA Headquarters. Because five major Shuttle disciplines derive their source of data from the wind tunnel testing process (aerodynamics, aerothermodynamics, airloads, structural dynamics, and stage separation), each discipline was represented on the panel by an associated subsystem manager. The SSV development covered a wide range of flight conditions throughout the Mach number range. Furthermore configuration modifications would result from the wind tunnel data analyses. For these two reasons it was decided to include a representative from each of the major U.S. wind tunnel testing complexes on the committee. These complexes were the NASA Langley (LaRC), Ames (ARC), and Lewis Research Centers (LeRC), and the Air Force Arnold Engineering Development Center (AEDC). The representatives were helpful not only in sharing similar test experiences, but they were also familiar with the capability and daily status of their respective facilities. They would be able to efficiently coordinate those tests scheduled at their facilities.

In September 1972, an organizational meeting of the SSECP-WTP was held at JSC to review with the prime contractor, Rockwell International, their overall estimates of the anticipated test program. Additionally the plans for the first 6 months of this program were reviewed in detail. Also guidelines for monitoring the overall test program were established. As a result the following procedures were adopted.

a. The wind tunnel coördination panel would meet quarterly until such a time that the panel would no longer serve its original purpose.

b. At each of these meetings Rockwell would provide a status of the major disciplines regarding what had been learned over the past several periods. Those issues would be addressed in the next segment of the program.

c. Rockwell would also review their master wind tunnel program regarding each proposed test to cover a 2-year period broken down as follows:

1. In detail, for the upcoming 6 months.
2. In general, for the following 6 months on a monthly basis.
3. A broad estimate for the subsequent 12-month period on a quarterly basis.

d. Each near-term proposed test, supported with a pretest run schedule, would be reviewed by the panel. The objectives, testing techniques, and facility utilization were to be reviewed.

e. Having satisfied the panel as to the test objectives, the facility representatives would provide tentative commitments from their facility management to support these panel-approved tests.

f. Before each test, the proposed run schedule would be required to have the concurrence of the appropriate JSC subsystem manager. The manager would indicate to the appropriate facility representative that the run schedule had been prioritized and minimized. The manager would also indicate whether the current data requirements of the SSV program had been met. Any facility disagreement with the test or run schedule would be negotiated between the JSC representative and the facility representative. Fig. 6.1 shows the flow of this chain of events.

g. The Rockwell onsite test engineer would have the responsibility for the real-time direction and run scheduling within the scope and objectives of the agreed-upon test. Any JSC inputs/modifications, to the conduct of the test, would be through the Rockwell test engineer.

h. Any major changes to the scope or objective of an approved test would be coordinated between the appropriate subsystem manager and the Rockwell analysis engineer, before reapproval. This reapproval would then be relayed to the facility representative and to the onsite test engineer.

i. Periodically, both the SSPO and/or the appropriate project office(s) would be briefed on the status of the test program and the analysis of the results.

During the time that the panel was active, September 1972 to September 1976, the SSECP-WTP met 12 times. At each of these meetings, the format was basically the same. Rockwell presented their proposed test program for review. It also presented comments on whether the data from previous testing were adequate and if any model modifications and/or additional testing were required. Rockwell also presented the overall status of the configuration development emphasizing the gathering of adequate data defining the baseline vehicle characteristics. MSFC presented similar information relative to the ET and SRB development. During the course of these presentations, problem areas, other then the "mainstream" vehicle characteristics, were identified. Action was assigned to one or more of the panel representatives for further analysis and resolution.

Many of the action items required major studies requiring technical specialists and appropriate testing facilities to provide timely data for analysis. As MSFC was primarily responsible for developing the ET and the SRB recovery system, it expended a large effort in defining the unsteady flow field of the mated vehicle. It also performed detailed interstage analyses and plume/support-interference studies. LaRC, with its long experience in aeronautics and large number of available facilities, provided the following major contributions to the SSV program. Included was the "fine-cut" stability and control analyses for the Orbiter. This information was used as the primary data source by the contractor for the final configuration analyses. Other LaRC studies included Orbiter center-of-gravity expansion capability, supersonic Orbiter hysteresis characteristics, RCS interaction effects, dynamic stability characteristics (Orbiter, integrated vehicle and ferry configuration), ALT/support-interference effects (base drag), launch vehicle drag reduction, tile roughness effects, and real-gas effects. LaRC heating studies covered boundary-layer transition; surface heating, and flow phenomena techniques (phase change coatings, oil flow, and electron beam); and interference heating. The ARC, although primarily involved in conducting the major segment of the development testing, did provide analytical support in the areas of Orbiter heating, aeroelasticity, static plumes, and aeronoise definitions.

The following paragraphs detail the major discussion items at each of the SSECP-WTP meetings. Reference to configurations should be correlated with the definitions given in the section "Configuration Evolution" or by using fig. 3.1.

SSECP-WTP Meeting No. 1

This first meeting was held at the end of October 1972, just before the configuration changes were finished for the ATP vehicle to form the PRR configuration.

<u>MSFC test requirements for accepted and proposed shuttle tasks.</u> - Seven tests were requested to support SRB entry analyses: four for ET and SRB ascent loads, one for the ET entry, one for lift-off aerodynamics, and one for launch vehicle plume simulation. Rockwell reviewed their required support contribution and concurred with five of the tests. They postponed the decision on the remainder until the next meeting.

<u>Space Shuttle aerodynamic design data base (ADDB).</u> - During the Phase B, B', and B" contractor studies, JSC formulated a document to serve as a single source of data for all aerodynamic-related studies. In addition to the basic vehicle and control surface aerodynamic data, related items such as configuration geometry definition, control surface deflection capability, etc. were included. This document guided Rockwell in producing the Space Shuttle ADDB. As one of the JSC supporting tasks, the data were also to be put on magnetic tape for the user community. Tight control ensured that all studies used this same source of "official" SSPO aerodynamic data from the ADDB's. This would avoid conflicting study conclusions that would occur if different sets of data were to be used (refs. 14 through 39).

<u>Wind tunnel test procedures.</u> - Procedures that were announced at the panel organizational meeting were changed. The pretest meetings were the responsibility of Rockwell and the facility. JSC would intervene only when conflicts arose. Also test data results were to be furnished to both Rockwell and JSC simultaneously.

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Plume testing. - Recent tests indicated that loads testing should include plume effects. Planned tests were revised to accommodate this need. In addition, a study was initiated to define a SRM nozzle shape that would minimize the plume effects on the vehicle aerodynamics.

Control effects tests. - A JSC study was presented that showed how the number of control surface settings (elevon) tested could be reduced and still give reliable results.

Air breathing propulsion system testing. - Because of the recent decision to eliminate the ABPS for entry (to be used for ferry only) these planned tests were switched to a more economical facility.

Reynolds number/ablation effects. - LaRC accepted a task to study the effects of Reynolds number in conjunction with ablation-caused surface roughness and shape. In addition, they would study TPS tile (waffle) effects with Reynolds number variations.

Test coordination. - Review and test revisions were accomplished in the areas of control surface flutter, panel flutter, aeronoise and RCS/flow interaction.

SSECP-WTP Meeting No. 2

This meeting was held in February 1973. The Orbiter configuration 2A, the "light-weight Orbiter," had just undergone preliminary definition. Several months of tests and analyses were required to confirm the configuration revisions. Tests identified by Rockwell for the near term included Orbiter Vehicle 2A preliminary stability and control, drag and ground effect ABPS studies, then Orbiter detailed stability and control. Also, Integrated Vehicle plume effects tests for stability and control were included. Orbiter loads tests were conducted to address design point conditions which were integrated into the structural analysis modal model. Separation tests had just been completed for the PRR configuration so only Orbiter abort separation tests were scheduled. These subjects also were discussed.

Plume testing review. - Plume testing requirements for the nominal ascent phase of flight, including separation and flight at hypersonic speeds, needed to be defined. This is so that the impact on facility hardware requirements could be made.

MSFC support activities. - The MSFC studies included the generation of launch heating environments, ET deorbit motor location effects, and protuberance effects. MSFC was also planning an ascent plume technology program to address both analytical and experimental plume simulation technology as related to the Shuttle. The objective was to provide an economical and efficient means for simulating SSME plume effects during the ascent phase. The program elements included a simulation technique evaluation that would qualify air simulation for known variable Γ gaseous plumes. It also included a Reynolds number (Re) evaluation to determine the quantitative effect of free-stream Re on plume-induced flow separation and a simulation evaluation to qualify air simulation using a hybrid (hot SRB, cold SSME plumes) model.

LaRC dynamic stability tests. - LaRC initiated a dynamic stability program to complement the one planned by Rockwell. Progress would be coordinated through the SSECP-WTP with JSC responsible for configuration definition.

SSECP-WTP Meeting No. 3

This meeting was held in May 1973. The short-lived Orbiter Vehicle 3 was about to become Vehicle 4. Rockwell presented results of some of their analytical prediction techniques for the Orbiter. Also presented was the test plan required to develop the Orbiter Vehicle 4 aerodynamics using existing Vehicle 2A and 3 models. Integrated Vehicle testing was planned to obtain parametric data (ET nose shape, Orbiter incidence, SRB location, attach structure detailed shape and Orbiter distance from the ET). Testing also gave Reynolds number effects, plume effects (location of SRB exit plane) and separation proximity aerodynamics. Because of the complicated test models required for air loads (distributed pressures) the model configurations would go directly from Vehicle 2A to 4.

In other actions, Rockwell was requested to plan RCS/flow interaction tests at hypersonic speeds and LaRC was requested to support sonic boom testing coordinated by a team from the ARC, MSFC, and JSC.

SSECP-WTP Meeting No. 4

The fourth meeting of the Wind Tunnel Panel was held in August 1973. Vehicle 4 was fairly well established. An extensive status report was given by Rockwell that reflected the Shuttle Requirements Review (SRR). As of this data 69 tests had been run in 6805 test hours using 17 facilities and 27 models. The estimate for the total program was 316 tests, 26 thousand hours, 31 facilities, and 69 models. The upcoming testing period would concentrate on the Orbiter PDR configuration due for review in February of 1974. Basic issues to be addressed were

a. Verification of basic stability and control capability.

b. Establishment of control surface effectiveness (off-nominal conditions and deflected surfaces for aileron/elevon and rudder/speedbrake combination settings through the complete Mach range).

c. Base sting interference and main rocket engine nozzle installation effects (wing tip or vertical tail extension stings used in conjunction with base dummy stings).

d. Configuration build-up (component on/off) effects.

e. Control surface hinge moments.

f. Vertical tail panel loads.

g. Reynolds number and viscous interaction effects.

h. ABPS location/configuration effects (ferry conditions).

i. Data tolerance level definitions (comparison of model/model, facility/facility, various model scales, etc.).

For the ascent vehicle the issues to be addressed were the definition of power-on base drag and continued ET nose shape effects. Also addressed were the protuberance and attach structure effects, booster separation rocket effects at long distances, and off-nominal relative attitudes. The airloads testing that was required concerned the issues of base pressures for a flared rudder, Orbiter/ET attach fitting simulation, additional rudder deflections, chord with denser pressure tap distribution, pressure taps to measure venting characteristics, jet exhaust effects on pressures, and pressure distributions on the ET and SRB's. Other discussion considered the abort testing requirements where only abort-to-once around (AOA) and RTLS conditions were to be tested, SRB/ET test requirements, and an addition to the wind tunnel test coordination procedure (biweekly teleconferences were initiated to allow premeeting test approval and definition of problem areas).

SSECP-WTP Meeting No. 5

The fifth panel meeting was held in November 1973. The Aerodynamics Subsystem Manager reviewed the separate milestones for the Orbiter Project and the Space Shuttle Program (Integrated Vehicle). Orbiter Vehicle No. 1 (OV-101) would be used in the ALT Program and Orbiter Vehicle No. 2 (OV-102) would be used in the Orbital Flight Test (OFT) program. Each had separate management reviews. In addition, delta PDR's and delta CDR's were set for OV-101 approximately 6 months after the scheduled PDR and CDR. The Aerodynamics Subsystem Manager established an "Aerodynamics PDR" to accomplish two objectives. The first was the documentation of the source and the analysis of the aerodynamic data and methods in the "Aerodynamics Substantiation Report." The second objective of this review would be to establish an aerodynamics verification plan, using flight and wind tunnel data, that would increase the confidence in the design data.

Orbiter aerodynamic issues for Vehicle 4 added since the last panel meeting were the effectiveness of vertical fins (increased yaw stability) and the effect of increasing the wing leading-edge radius on stability and control. Ascent aerodynamics tests continued on separation/plume effects including RTLS separation of slightly higher speeds. Airloads issues were the acquisition of detailed pressure distributions with RCS plume simulation effects, for prelaunch, for asymmetric effects such as attach fittings and the ET external feedline, for ABPS effects, and for powered SRB separation. SRB concerns were Reynolds number effects (chute deployment altitude and attitude), high Mach/high stability (chute altitude); strakes, and sting interference (accuracy of data). The plume technology tests by the MSFC included analysis of air/CF_4 testing, plans to test at higher chamber pressure ratios, SRB hot gas test activities, and solid body plume testing for Reynolds number effects. Aerothermodynamics testing was being done to address boundary-layer transition effects, specific heat ratio effects, and detailed heating definitions for a ventral fin, the wing leading edge and TPS gaps.

SSECP-WTP Meeting No. 6

The sixth wind tunnel meeting was held in March 1974, directly following the OV-101 PDR in February. The results of the PDR presented by JSC concluded that OV-104 will meet the aerodynamic ground rules and the identified requirements. The Orbiter Vehicle testing requirements for the newly designated Vehicle 5 included viscous interaction effects on hypersonic stability and control. The definition of aerodynamic data tolerances was improved. Additional stability and control data was required. Sting interference effects, TPS simulation, and air data sensor (ADS) calibrations were added. Integrated Vehicle requirements covered nominal and RTLS staging (build-up data and rudder/elevon hinge moments). Also covered were power-on base drag definition (cold gas, base pressures, wing/vertical tail bending moments, and hot/cold gas comparisons). RCS effectiveness (hot/cold gas for RTLS) and mated vehicle aerodynamics (second stage, sting effects, SRB nose/skirt parametrics) were included. Airloads test requirements were discussed. Rockwell was requested to combine the objectives of the planned airloads/force test, plume effects test, and the base drag assessment test. A long-term schedule for structural dynamics showed the need for TPS development tests to support the planned TPS PDR in November 1974. Aerothermodynamic test requirements covered the effects of protuberances, penetrations and gaps on heating, and RTLS abort heating (high α's). Plume tests planned included development base convective heating and pressure. These tests also included plume/boundary-layer interactions, "creep" heating and pressure, RTLS abort heating and pressure, and launch pad/Integrated Vehicle interaction heating. Entry heating tests were planned for updated overall configuration heating and pressure, wing leading-edge heating, and SSME nozzle heating. SRB planned tests were to obtain updated configuration stability and heating/pressure distributions as well as drogue deployment feasibility data. Similar testing was to be done for the ET. In addition, a test to define SRB sonic boom characteristics was planned.

SSECP-WTP Meeting No. 7

This meeting was held in July 1974. Because of uncertainty regarding the SSV 1975 Fiscal Year budget the panel meeting had been delayed for a month. During this period JSC had requested Rockwell to organize their proposed testing to meet a "minimum requirements" program for the Vehicle 5 configuration. Proposed Orbiter aerodynamics testing addressed the evaluation of recent design changes (elevon gaps, OMS pods, differential elevons, and elevon flapper doors). Also addressed was the determination of viscous interaction effects on hypersonic stability and control. Inboard/outboard split elevon effectiveness, RCS simulation improvement, and testing required to support TPS simulation was also discussed. Integrated Vehicle tests were proposed to obtain basic aerodynamics for Vehicle 5. This included elevon/rudder/bodyflap hinge moments, wing root bending and torsion, and plume effects (on base pressures, nozzle loads, elevon characteristics). Separation testing was planned for nominal conditions at high α's and β's, and for RTLS conditions with RCS jet simulation. SRB alone tests covered entry stability, nozzle hinge moments, Reynolds number effects, and pressure distributions for both venting studies and load calculations. ET tests would cover protuberance loads and sonic boom characteristics. Airload tests were to be conducted for pressures near the Orbiter nose and main wheel well (thermal blanket survival assessment). Flow visualization and wake mixing data, exhaust plume effects, elevon deflection effects on wing pressures, and element pressures (Orbiter, ET and SRB mated) were also tested. Structural dynamics testing was for aeronoise effects on the TPS (structural panel tests), flutter with TPS (structural panel), and flutter for the wing/elevon-fin rudder components. Aerothermodynamics tests concentrated on TPS tile gaps with pressure gradients. Thin film gauge instrumented Orbiter/ET attach structure for the ascent phase and surface roughness (paint) and canopy/forebody heating rates for entry conditions was also tested. For the carrier program, testing was about to begin. The Boeing Company had been awarded the contract the previous month. Some preliminary design support tests had been done leading up to carrier ATP. Now the detail design test requirements had to be done in preparation for carrier PDR in November 1974. Near-term testing included mated vehicle stability and control for takeoff. Cruise and landing (including effects of Orbiter incidence, Orbiter position, tailcone shape and carrier vertical tail modifications) were tested. Separation configuration development (Reynolds number effects and separation matrix definition) and tailcone configuration definition were tested as well.

At this stage of the program, the major configuration changes had been made. The bulk of the future tests would be for configuration refinements, data verification, and contingency issues. It was decided therefore, to reduce the quarterly panel meetings to three per year after the next meeting.

SSECP-WTP Meeting No. 8

The eighth meeting of the panel took place in October 1974. Program schedules were presented and discussed in terms of events related to the panel subsystem manager's responsibilities. Fabrication of most of the Orbiter structure for OV-101 was completed and assembly had begun.

Major aerodynamic issues for the Orbiter were the following:

a. Increased drag caused by the shortened (blunted) OMS pods

b. Unaccounted wave drag caused by TPS tile steps

c. Initiation of a test program for the entry ADS probes

d. Decrease in pitching moment caused by viscous-interaction effects

e. Corrections for base sting and SSME nozzle effects

f. Influence of RCS "RT" (product of gas constant and temperature) scaling and Mach effect on entry RCS simulation

g. Reduced rudder effectiveness caused by air leakage through the rudder hinge line gap.

Integrated Vehicle aerodynamic issues discussed were Vehicle 5 power-on base drag test results (higher drag level than the latest data book). Also discussed were the Vehicle 5 power-off forebody drag (recent tests show higher drag) and elevon deflection schedules for hinge moment actuator load relief. For the carrier program, a requirement to land with the tailcone had been added. Therefore a test series was initiated to address this configuration. SRB testing covered Reynolds number effects, updated configuration stability and control, and nozzle hinge moments. ET tests included Orbiter/ET fairing optimization, Reynolds number effects, dynamic stability verification, and plume simulation effects. Entry heating tests were planned to obtain the effects of forebody boundary-layer transition and surface roughness, surface seals and cavities, leading-edge radius, and TPS gaps. Integrated Vehicle heating tests covered updated configuration heating distributions, protuberance effects, and base convective heating/pressures. Structural dynamics tests were to be done for wing and vertical tail flutter boundary updates.

SSECP-WTP Meeting No. 9

This meeting was held in May 1975. Orbiter aerodynamic concerns were trim capability near Mach 5, loads exceeding the transonic elevon hinge moment limits, Reynolds number effects, ALT airloads, and continued air data probe calibrations. The Integrated Vehicle testing plans were for RTLS separation and airloads with plumes (hinge moments, venting pressures). The carrier aerodynamic issues were strut (Orbiter) effects, launch separation, captive vehicle airloads, and tailcone off buffet alleviation. Orbiter heating tests and analyses addressed the establishment of smoothness criteria, preventing early flow transition, and elevon dynamic seal heating. Ascent heating concerns were Vehicle 5 heating and pressure distributions and plume effects on base heating at high altitudes. Structural dynamics testing was on panel flutter with and without TPS, aeronoise during ascent, and rigid Orbiter tests to predict buffet onset (including canopy effects).

SSECP-WTP Meeting No. 10

The tenth panel meeting was held in October 1975. Orbiter aerodynamics concerns to be addressed included viscous interaction and real-gas effects. Also discussed were effects of new outer mold lines (OML) caused by TPS thickness redefinition, ALT vehicle tailcone data deficiencies, and air data probe (forebody model) calibrations. Integrated Vehicle aerodynamic issues were force and moment element data with an updated elevon deflection schedule. Other issues were RTLS abort separation with force and aft RCS effects, AADS feasibility, airloads update, and effects of protuberances and attach hardware. Entry heating tests involved the definition of boundary-layer transition. For the Integrated Vehicle, heating tests continued on Vehicle 5 heating and pressure distributions, and protuberance effects. Structural dynamics concerns were for Orbiter Nomex felt panel flutter.

SSECP-WTP Meeting No. 11

A review of the remaining test program in this March 1976 meeting indicated that because of the advance approval management procedure that was used, the Wind Tunnel Panel's role was essentially completed. The panel decided to dissolve itself after the next meeting and transfer the management of any future tests to the respective subsystem managers.

The OV-101 vehicle had been assembled and was approaching the Shuttle ALT CDR. The OV-102 vehicle OML definition had been signed off, fabrication had started, and assembly was 4 months off. Orbiter aerodynamic issues concerned

a. Verification of the OV-102 lines at subsonic and transonic speeds

b. Nonlinear aerodynamic control surface characteristics (LaRC-supported)

c. Predictions of aeroelastic effects

d. Evaluation of the ALT ADS calibrations (side probes and nose boom) and OV-102 ADS test plans

e. Aerodynamic loads on landing gear (which are gravity operated) struts and doors.

f. Tailcone on data (ground effects, airloads, hinge moment effects, and test support system tares).

Integrated Vehicle studies concentrated on continuation of tests on elevon relief (to maintain both wing root loads and elevon hinge moments within design limits), AADS feasibility, and RTLS separation. Carrier aerodynamic issues were verification of the data base for mated vehicle launch and cruise configuration, take-off and landing configuration, and separation. Structural dynamics tests continued on the effects of local shocks and aeronoise on the TPS. Integrated Vehicle and Orbiter heating tests continued to address localized heating and configuration updates.

SSECP-WTP Meeting No. 12

At this last official meeting of the panel in September 1976, it was requested that each remaining test be identified to the appropriate subsystem manager in detail and in writing. The subsystem managers would then review these test data requirement packages (test description sheets for the remainder of the program and detailed run schedules through July 1977). An assessment of the occupancy hours would be made by the facility representatives. Then a memorandum would be sent from the JSC Director requesting that these tests be conducted. It was estimated that the current percent of SSV test completion was 83 percent for aerodynamics, 62 percent for heating, and 71 percent for structural dynamics. Test hours estimated for completion were 40,700 with 29,900 having been accomplished, or 73 percent. Projected testing that remained would be primarily in the areas of verification testing and "as-built" configuration testing. The purpose of verification testing was to furnish data for the design data uncertainties analysis. The "as-built" tests would address data gaps, configuration updates and discrepancies, and data discrepancies. These latter tests would be in the highest quality facilities available using completely new high fidelity models.

Remaining Orbiter aerodynamic testing addressed the entry ADS, refinement of stability and control and hinge moments, aeroelastic effects (wing and vertical tail), hypersonic viscous interaction/real-gas effects, and various off-nominal flight conditions. Integrated Vehicle tests would cover refined plume effects, the AADS calibration, and contingency airloads. Orbiter and Integrated Vehicle heating addressed configuration updates and localized heating regions. Structural dynamics tests were to be conducted on updated configurations for flutter/buffet, panel (TPS) flutter, oscillatory pressure effects, hypersonic bodyflap buzz, and ground winds data verification. Tests were also to continue through 1976 on SRB recovery conditions and for udpated SRB and ET configuration aerodynamics, structural dynamics, and heating.

SSECP-WTP Special Meeting

In November 1977, a special meeting of the panel was held to update the remaining test program through the First Manned Orbital Flight (FMOF). Contingency and postflight verification tests were scheduled and were to be canceled as soon as it was determined they were not necessary.

7. WIND TUNNEL TEST PROGRAM

As previously discussed, obtaining the aerodynamic, aerothermodynamic, and structural characteristics for the design and development of the SSV configurations necessitated the formulation and execution of an extensive wind tunnel testing program. To minimize costs, plans were made to make maximum use of NASA test facilities and to fabricate multipurpose wind tunnel models. The contractor Phase C/D wind tunnel test program, which began in September 1972, was formally completed in September 1983.

In the formal Rockwell proposal, the estimated wind tunnel test program consisted of 18,900 hours for the Orbiter and 7,100 hours for the mated launch vehicle and the elements for a total of 26,000 hours. During the test program, several management decisions required major configuration changes. Changes such as Orbiter sizing, Orbiter/carrier concept, launch vehicle operational concepts, etc. (fig. 3.1 and 3.2 for the major design evolution characteristics). This resulted in a substantial increase over the initially proposed number of test hours. Overall the prime contractor-utilized approximately 46,500 wind tunnel test hours in the Phase C/D time period. Table 7.1 presents a summary of the test hours per major configuration, for the various disciplines, together with the number of supporting models. As a projection of the cost for the test program, the facility hours and models are shown with their estimated costs. Fig. 7.1 illustrates the occupancy hours for the various configurations per discipline, in comparison with similar large-scale test programs.

In addition to the Rockwell Phase C/D wind tunnel test program, MSFC and LaRC major-supporting test programs during this time amounted to approximately 20,000 hours. The MSFC, having direct responsibility to the SSV program, expended approximately 7,000 hours performing in-house SSV technology tasks. JSC conducted approximately 1,800 hours of tests, primarily in direct support of Orbiter aerodynamics.

Wind tunnel facilities used in the test program were chosen based on the tunnel's capability to simulate the required test conditions. However, the selection was also based on the facilities cost and efficiency. For this reason, nearly 71 percent of the test program was conducted in NASA facilities. Specifically, 41 percent was done at ARC, 17 percent at MSFC, 13 percent at LaRC, and 1 percent at LeRC.

To assist in the test program management and coordination, a test coding system was developed. The tests were divided into groups each representing the particular organization which was directly responsible for that test (i.e. Rockwell, LaRC, MSFC, etc.). Each test was given an alpha/numeric identification code. For the Rockwell tests, the first of the two alpha characters indicated whether the tests were for the Orbiter (O), Integrated Vehicle (I), carrier aircraft (C), external tank (T), or solid rocket booster (S) configuration. The second alpha character denoted the area of discipline to be evaluated: aerodynamics (A), heating (H), or structures (S). The numerical characters represented the chronological order of the tests. It should be noted that the airloads and the ascent phase separation aerodynamics are listed under "A". Structures tests are predominately structural dynamics tests (as opposed to airloads tests). Also the heating test program included some pressure distribution testing that was done simultaneously (at the identical test conditions) with the heating tests to ensure data analysis compatibility.

For those support tests conducted by the various NASA centers, the first alpha character was changed from the above description and was used in identifying the center as LaRC (L), MSFC (F), ARC (A), and JSC (M), with the remainder of the identification code the same as previously explained.

A summary of the total test program is given in table 7.2 and the detailed lists of tests for the Orbiter, Integrated Vehicle, carrier vehicle, ET and SRB's are contained in appendices A1 through A5, respectively. Each table gives the test number (or ID), the test schedule, the occupancy hours that were estimated and that were charged by the facility (actual "fan-on" hours), the actual number of runs conducted, the model reference designation (see "Configuration Evolution" section) and model ID (Refer to "Wind Tunnel Test Models" section), the facility used (Refer to "Wind Tunnel Facilities" section) and the facility-designated test number, and the DATAMAN document number and status (Refer to "Documentation" section). Not shown in the test program listings is approximately 660 hours representing 12 Orbiter ADS probe-alone tests.

TABLE 7.1. - FACILITY HOURS AND ESTIMATED COST FOR THE PRIME CONTRACTOR

		AERODYNAMICS	HEATING	STRUCTURAL DYNAMICS	TOTAL
ALT/CARRIER	HOURS	3,500	—	400	**3,900**
	COST	5.3$\overline{m}$	—	0.6$\overline{m}$	**5.9$\overline{m}$**
ENTRY	HOURS	17,000	5,400	2,500	**24,900**
	COST	25.5$\overline{m}$	8.1$\overline{m}$	3.7$\overline{m}$	**37.3$\overline{m}$**
ASCENT	HOURS	10,300	6,000	900	**17,200**
	COST	15.4$\overline{m}$	9.0$\overline{m}$	1.4$\overline{m}$	**25.8$\overline{m}$**
FACILITY TOTAL	HOURS	30,800	11,400	3,800	**46,000**
	COST	46.2$\overline{m}$	17.1$\overline{m}$	5.7$\overline{m}$	**69.0$\overline{m}$**
MODEL TOTAL	NO.	45	34	21	**100**
	COST	12.6$\overline{m}$	9.0$\overline{m}$	4.9$\overline{m}$	**26.5$\overline{m}$**
ESTIMATED WIND TUNNEL PROGRAM COST	**TOTAL**	**58.8$\overline{m}$**	**26.1$\overline{m}$**	**10.6$\overline{m}$**	**95.5$\overline{m}$**

NOTE: $\overline{m}$ = MILLION DOLLARS

TABLE 7.2. - PHASE C/D SSV COMPILATION OF THE WIND TUNNEL TESTING TOTAL ACTUAL PROGRAM

AERODYNAMICS			HEATING			STRUCTURES			TOTAL	
TEST	HOURS	NO. OF TESTS	TEST	HOURS	NO. OF TESTS	TEST	HOURS	NO. OF TESTS	HOURS	NO. OF TESTS
OA	16988	149	OH	5387	79	OS	2487	40	24862	268
IA	10275	104	IH	5936	46	IS	854	10	17065	160
CA	3470	23	CH	0	0	CS	425	6	3895	29
RI/TOT	30733	276	RI/TOT	11323	125	RI/TOT	3766	56	45822	457
SA	3494	27	SH	344	4	SS	1576	15	5414	46
TA	472	5	TH	72	2	TS	0	0	544	7
MA	3030	28	MH	75	2	MS	0	0	3105	30
FA	4465	24	FH	324	6	FS	824	6	5613	38
LA	10231	128	LH	0	0	LS	0	0	10231	128
AA	568	5	AH	0	0	AS	0	0	568	5
TOTAL	52993	493	TOTAL	12138	139	TOTAL	6166	77	71297	711

8. WIND TUNNEL TEST FACILITIES

Many facilities were required to simulate the particular flight conditions that would provide the design of the many SSV configurations. These data had to be accumulated over the entire range of flight conditions that would be encountered from launch through entry and then some. The selection of a tunnel, for any particular test was based on the tunnel's capabilities to simulate required flight conditions, principally Mach and Reynolds number. However, selection was also based on the facility, convenience, and operational economy when there were possible alternate choices.

There were three primary types of wind tunnels used.

a. Continuous flow tunnels. These tunnels permit uninterrupted testing until all required data are obtained on a particular configuration. Models and test conditions of Mach number, Reynolds number, and angle of attack and sideslip are set up and the air (or test medium) is recirculated until run completion.

b. Intermittent tunnels (blowdown tunnels). These tunnels have an operating time from several seconds to a few minutes. They have storage tanks charged with pressurized air that is suddenly released, and data are taken over a short blowdown time span.

c. Impulse tunnels. These tunnels are designed for the simulation of high Mach number values. They have a very short run time, in the order of milliseconds. Instrumentation of the models for this type of tunnel must have high response for the instantaneous recording of the necessary data.

The capability of the various facilities to match the combined Reynolds number/Mach number flight conditions is given in fig. 8.1(a) for ascent and fig. 8.1(b) for entry. In the high hypersonic regime (Mach number above 5) the viscous parameter, $\bar{V}'_{\infty}$, was used as the correlation parameter rather than Reynolds number (see fig. 8.2). Facility information for the SSV test program is summarized in table 8.1 grouped by speed category. The facility name, type, test section size, Mach number range, Reynolds number range and typical model scales is given.

The choice of facility for any given test was determined by selecting the one which could approximate flight conditions. Flight conditions were approached through control of the geometric similarity and scale of models (properly instrumented) and tunnel control of

a. Reynolds number, which is the ratio of the test medium's inertia force to the viscous force. The similarity between a model and prototype is realized when the dimensionless Reynolds number for the model equals the Reynolds number for the prototype. In some instances, it was the objective of a particular wind tunnel test to show that the effect of varying the Reynolds number is a negligible factor. Therefore some tests were conducted at several Reynolds number levels.

b. Mach number, which is the ratio of the relative velocity of the vehicle to the speed of sound in the medium.

c. Angle of attack and sideslip, which is the attitude of the model in relation to the free-stream velocity vector in the pitch and yaw planes, respectively.

d. Other parameters such as thrust coefficients, aeroelastic frequencies, separation distances, etc. that had to be simulated for specific tests.

Because of the impossibility of obtaining all of the required data in one facility, it was necessary to use combinations of facilities and models to obtain data over the desired range of flight conditions.

Additionally, there were no tests or combination of tests that could account for all variables such as Reynolds number, Mach number, angle of attack, model size, wind tunnel, or balance choice that could be made in such a manner as to provide all desired data for the determination of aerodynamic characteristics. Therefore, testing was directed to those areas considered most critical to the vehicle's design. With the limited data obtained it was possible to predict, or extrapolate, the aero/thermo/structural characteristics in the untested areas.

The complete phase C/D wind tunnel program is grouped by facility designation and presented in appendix A. The in-house facility test number is given followed by the SSV test designation, test date, hours, runs, model configuration reference and ID, and documentation number.

TABLE 8.1 - LIST OF SOME OF THE MAJOR SUPPORTIVE TEST FACILITIES

FACILITY NAME	TYPE	TEST SECTION	MACH NUMBER	REYNOLD'S NUMBER X 10**6/FT	MODEL SCALE RANGE
SUBSONIC					
ARC 12-FT PRESSURE TUNNEL	CONTINUOUS	11.3 BY 11.3 FT	0.0 - 0.9	0.0 - 9.0	1.5 - 5.0
ARC 40- BY 80-FT SUBSONIC TUNNEL	CONTINUOUS	40.0 BY 80.0 FT	0.0 - 0.3	0.0 - 2.1	33.33
TBC V/STOL TUNNEL	CONTINUOUS	20.0 BY 20.0 FT	0.0 - 0.1	0.0 - 2.5	4.0
LARC LOW TURBULENCE PRESS. TUNNEL	CONTINUOUS	7.5 BY 3.0 FT	0.1 - 0.4	0.6 - 15.0	2.0
LRC V/STOL	CONTINUOUS	14.5 BY 21.5 FT	0.0 - 0.3	0.0 - 0.5	4.0
LTV 15- BY 20-FT SUBSONIC TUNNEL	CONTINUOUS	15.0 BY 20.0 FT	0.0 - 0.1	0.6 - 0.5	5.0
RI 7- BY 10-FT LOW SPEED TUNNEL	CONTINUOUS	7.0 BY 10.0 FT	0.0 - 0.3	0.0 - 2.0	2.0 - 4.0
TAMU 7- BY 10-FT SUBSONIC TUNNEL	CONTINUOUS	7.0 BY 10.0-FT	0.0 - 0.3	0.0 - 1.9	4.0
UNIV. OF WASH. 8- BY 12-FT TUNNEL	CONTINUOUS	8.0 BY 12.0-FT	0.0 - 0.3	0.0 - 1.8	4.0
TRANSONIC					
ARC 11- BY 11-FT TRANSONIC TUNNEL	CONTINUOUS	11.0 BY 11.0 FT	0.5 - 1.4	1.7 - 9.4	2.0 - 5.0
ARC 2-FT TRANSONIC TUNNEL	CONTINUOUS	2.0 BY 2.0 FT	0.6 - 1.4	0.5 - 8.7	----------
ARC 14-FT TRANSONIC TUNNEL	CONTINUOUS	13.5 BY 14.0 FT	0.6 - 1.2	2.8 - 5.2	1.25 - 3.0
AEDC 4-FT PROPULSION TUNNEL	CONTINUOUS	4.0 BY 4.0 FT	0.2 - 1.3	0.2 - 7.0	0.6
AEDC 16-FT TRANSONIC TUNNEL	CONTINUOUS	16.0 BY 16.0 FT	0.2 - 1.6	0.2 - 6.0	2.0 - 3.0
TBC TRANSONIC WIND TUNNEL	CONTINUOUS	8.0 BY 12.0 FT	0.2 - 1.2	0.0 - 4.0	4.0
CAL 8-FT TRANSONIC TUNNEL	CONTINUOUS	8.0 BY 8.0 FT	0.0 - 1.4	0.7 - 7.0	1.5 - 2.0
LARC 8-FT TRANSONIC PRESS. TUNNEL	CONTINUOUS	7.1 BY 7.1 FT	0.2 - 1.3	0.1 - 6.0	1.5 - 2.0
LARC 16-FT TRANSONIC DYNAMIC TUNNEL	CONTINUOUS	16.0 BY 16.0 FT	0.2 - 1.6	0.0 - 9.7	2.0 - 4.0
LARC 16-FT TRANSONIC TUNNEL	CONTINUOUS	16.0 BY 16.0 FT	0.2 - 1.3	1.2 - 3.7	1.5 - 5.0
LTV HIGH SPEED WIND TUNNEL	INTERMIT	4.0 BY 4.0 FT	0.2 - 5.0	2.0 - 38.0	1.5

TABLE 8.1 - CONCLUDED

FACILITY NAME	TYPE	TEST SECTION	MACH NUMBER	REYNOLD'S NUMBER X 10**6/FT	MODEL SCALE RANGE
SUPERSONIC					
==========					
ARC 8- BY 7-FT SUPERSONIC TUNNEL	CONTINUOUS	8.0 BY 7.0 FT	2.4 - 3.5	0.5 - 5.0	2.0 - 5.0
ARC 9- BY 7-FT SUPERSONIC TUNNEL	CONTINUOUS	9.0 BY 7.0 FT	1.5 - 2.6	0.8 - 6.5	2.0 - 5.0
ARC 6- BY 6-FT SUPERSONIC TUNNEL	CONTINUOUS	6.0 BY 6.0 FT	0.6 - 2.2	0.5 - 5.0	1.5
AEDC TUNNEL "A"	CONTINUOUS	3.3 BY 3.3 FT	1.5 - 6.0	0.3 - 9.2	1.0 - 2.0
AEDC 16-FT SUPERSONIC TUNNEL	CONTINUOUS	16.0 BY 16.0 FT	1.5 - 2.4	0.2 - 2.5	1.0 - 2.0
LARC UNITARY PLAN TUNNELS	CONTINUOUS	4.0 BY 4.0 FT	1.5 - 4.6	0.4 - 8.0	1.0 - 2.0
LERC 10-BY 10-FT SUPERSONIC TUNNEL	CONTINUOUS	10.0 BY 10.0 FT	2.0 - 3.5	0.1 - 3.4	2.0 - 4.0
RI 7- BY 7-FT TRISONIC TUNNEL	INTERMIT	7.0 BY 7.0 FT	0.1 - 3.5	2.0 - 17.0	0.4 - 1.5
MSFC 14-INCH SUPERSONIC TUNNEL	INTERMIT	14-INCH DIA.	0.6 - 4.5	3.0 - 18.0	0.4
HYPERSONIC					
==========					
ARC 3.5-FT HYPERSONIC TUNNEL	INTERMIT	3.5-FT DIA.	5 & 7.5	0.3 - 7.4	1.0 - 1.75
AEDC TUNNEL "B"	CONTINUOUS	50-INCH DIA.	6 & 8.0	0.3 - 5.3	1.0 - 2.0
AEDC TUNNEL "C"	CONTINUOUS	50-INCH DIA.	10 & 12	0.3 - 2.4	1.5
AEDC TUNNEL "D"	CONTINUOUS	1.0 BY 1.0 FT	1.5 - 5	0.3 - 16.0	1.75.0
AEDC TUNNEL "F"	INTERMIT	25-INCH DIA.	7	4.5 - 45.0	1.0 - 1.75
CAL 8-FT HYPERSONIC TUNNEL	SHOCK	72-INCH DIA.	6	0.0 - 75.0	1.0.5 -'24
NSWC TUNNEL "#9"	SHOCK	5-FT DIA.	1	0.4 - 5.8	2.04
LARC 20-INCH MACH 6 TUNNEL	INTERMIT	20 BY 20 INCH	6	0.7 - 9.3	0.4 - 1.0
LARC VARIABLE DENSITY TUNNEL	INTERMIT	18-INCH DIA.	7	0.1 - 12.0	0.6.5 - 8.0
LARC CONT. FLOW HYPERSONIC TUNNEL	INTERMIT	31 BY 31 INCH DIA.	10,11,12	0.3 - 2.3	0.6 - 1.0
LARC 22-INCH HELIUM TUNNEL	INTERMIT	22.5-INCH DIA.	18 - 22	0.7 - 11.3	0.4
LARC HYPERSONIC NITROGEN TUNNEL	INTERMIT	19-INCH DIA.	19	0.2 - 1.3	0.4 - 0.6
LARC 20-INCH FREON TUNNEL	INTERMIT	20-INCH DIA	6	0.2 - 1.0	0.4 - 0.6
LARC 4-FT HYPERSONIC TUNNEL	INTERMIT	48-INCH DIA.	8 - 18	0.0 - 1.0	0.4

9. WIND TUNNEL TEST MODELS

Based on the range of flight regimes that were to be evaluated in developing the design of the SSV, a total of over 100 models were eventually required. The model scales range from 0.4 to 36 percent of the full-scale configuration. Exceptions to these scales were the full scale testing of specialized components such as control panels, TPS tile with gaps, etc. Model size was determined by the required testing parameters and the effective tunnel size for attaining flight simulation with minimum tunnel interference. The models were geometrically scaled and manufactured to extremely close tolerances to obtain accurate data. Later models incorporated finer details of external protuberances, surface roughness and indentations, as shown graphically in fig. 9.1.

Force models, which measure the three forces and three moments that define the overall performance and stability of the vehicle, were the primary source for obtaining the aerodynamics of the SSV configurations. In the early tests when basic force and moment testing was being done in the low speed facilities, wooden models were used. Soon after, for testing in the high speed/pressurized facilities, the models were made of aluminum and/or stainless steel. Ultimately, for hypersonic facilities, the models were constructed of heat-treated steel.

Heat transfer models were used to establish the flow field and the temperature distribution about the vehicle. The models were either made of a plastic-like material or they were stainless steel models. The former type models were used to define the flow patterns using oil-flow photographs. The stainless steel models were fitted with thermocouples (and/or calorimeters) to define the temperature distribution and stagnation heating for a range of trajectory conditions.

Pressure models were used to obtain detailed surface pressure distributions to assist in the airloads analysis and to design the various structural components of the SSV configurations. Pressure models were basically the same as the force models (and in some cases identical models). They were fabricated from aluminum and/or steel. The major differences, however, were the multitude of pressure ports located over the surface of the model and the required plumbing located in the interior of the model to measure all the pressures.

Aeroelastic models were used to measure the torsional, shear, and bending characteristics of a particular component. The elastic models were force models with the component to be evaluated (e.g. the vertical tail or wing) being replaced with an elastic component.

Flutter models were used to evaluate the dynamic characteristics of a particular component. Flutter models were usually made of balsa wood with appropriate stiffness. The flutter models are normally tested to destruction.

A summary of various models used is shown in table 9.1 with definition of the model identification (ID), configuration represented, major test discipline, scale, type measurements, and general comments.

Appendix A gives additional information, grouped by model. It includes the test the model was used in, test dates, test hours, number of runs, the facility used (and facility test number), and documentation information.

TABLE 9.1 - SUMMARY OF THE SPACE SHUTTLE WIND TUNNEL MODELS

MODEL NUMBERS	PRINCIPLE DISCIPLINE	MODEL SCALE	BASIC FUNCTION	GENERAL COMMENTS
1-0TS	AERODYNAMICS	0.004	FORCE & MOMENTS	MODIFIED PRE-ATP STABILITY & CONTROL MODEL
2-0	AERODYNAMICS	0.045	FORCE & MOMENTS	PRELIMINARY SUBSONIC STABILITY & CONTROL MODEL
3-0T	THERMODYNAMICS	0.006	PAINT	HEAT TRANSFER DISTRIBUTION MODEL
4-0T	THERMODYNAMICS	0.014	THERMOCOUPLES	THIN-SKIN ORBITER FOREBODY MODEL (CANOPY)
5-0T	THERMODYNAMICS	0.015	THERMOCOUPLES	THIN SKIN ORBITER FOREBODY MODEL (CANOPY)
6-0TS	AERODYNAMICS	0.015	FORCE & MOMENTS	DEVELOPMENT/STABILITY & CONTROL MODEL
7-0TS	AERODYNAMICS	0.019	PRESSURES:JET EFFECTS	COLD-JET PLUME SIMULATION ASCENT MODEL
8-0	STRUCT. DYNAMICS	0.046	AEROELASTIC:ORB/CARRIER	RIGID ORB/SCA CONFIG. WITH FLEX. 747 VERT. TAIL
9-0TS	AERODYNAMICS	0.0075	FORCE/PRESS.:SEPARATION	STABILITY & CONTROL SEPARATION MODEL
10-0TS	AERODYNAMICS	0.01925	FORCE GROUND WINDS	LAUNCH/LAUNCH PAD SIMULATION MODEL
11-0TS	STRUCT. DYNAMICS	0.04	PRESSURES	AERO NOISE MODEL
12-0	------CANCELLED, NOT BUILT			
13-0TS "F"	AERODYNAMICS	0.004	FORCE/PRESSURE	STABILITY & CONTOL MODEL
13-0TS "P"	AERODYNAMICS	0.004	PRESSURES	PRESSURE MODEL
14-0TS	AERODYNAMICS	0.019	FORCE/PRESS.: JET EFFECTS	COLD-JET PLUME SIMULATION ASCENT MODEL
15-0	THERMODYNAMICS	1.000	THERMOCOUPLES:HRSI TILES	FLAT PLATE: THIN SKIN TPS & PROTUBERANCES
16-0	AERODYNAMICS	0.0405	FORCE & MOMENTS	DEVELOPMENT SUBSONIC STABILITY & CONTOL MODEL
17-0TS	AERODYNAMICS	0.030	PRESSURE/FORCE	AIRLOADS MODEL
18-0	AERODYNAMICS	0.015	FORCE & MOMENTS	STABILITY & CONTROL MODEL

19-0TS	THERMODYNAMICS	0.0225	PRESSURE: JET EFFECTS	HOT-JET PLUME SIMULATION ASCENT MODEL
20-0	-------CANCELLED, NOT BUILT			
21-0T	THERMODYNAMICS	0.0175	PAINT: JET EFFECTS	RCS (VARIABLE NOZZLES) SIMULATION MODEL
22-0TS	THERMODYNAMICS	0.0175	THERMOCOUPLES	THIN SKIN TEMPERATURE MODEL
23-0	STRUCT. DYNAMICS	0.02	WING/ELEVON FLUTTER	PARTIAL WING/ELEVON MODEL
24-0	STRUCT. DYNAMICS	0.025	VERT./RUDDER FLUTTER	PARTIAL VERTICAL/RUDDER MODEL
25-0	THERMODYNAMICS	0.04	BASE TRANSDUCERS	HOT-JET PLUME ASCENT AFTERBODY MODEL
26-0TS	THERMODYNAMICS	0.01	PRESSURES	PRESSURE DISTRIBUTION MODEL
27-0	AERODYNAMICS	0.015	FORCE: RCS EFFECTS	STABILITY & CONTROL / RCS MODEL
28-0TS	-------CANCELLED, NOT BUILT			
29-0	THERMODYNAMICS	0.0175	CALORIMETERS	HEAT TRANSFER DISTRIBUTION MODEL
30-0	STRUCT. DYNAMICS	0.0125	STRAIN GAGES: LOADS	WING REFLECTION PLANE FLUTTER MODEL
31-0	THERMODYNAMICS	0.00593	PAINT	HEAT TRANSFER DISTRIBUTION MODEL
32-0TS	AERODYNAMICS	0.01	FORCE & MOMENTS	STABILITY & CONTROL MODEL
33-0	THERMODYNAMICS	0.00593	PAINT	HEAT TRANSFER DISTRIBUTION MODEL
34-0TS	AERODYNAMICS	0.004	FORCE & MOMENTS	STABILITY & CONTROL MODEL
35-0	THERMODYNAMICS	0.182	THERMOCOUPLES	FLAT PLATE: SHOCK GENERATOR MODEL
36-0TS	AERODYNAMICS	0.015	PRESSURES	RCS VENTING/PLUME MODEL
37-0T	THERMODYNAMICS	0.01	THIN FILM GAGES	HEAT TRANSFER DISTRIBUTION MODEL
38-0	THERMODYNAMICS	0.0058	PAINT	HEAT TRANSFER DISTRIBUTION MODEL
39-0	AERODYNAMICS	0.05	FORCE & MOMENTS/PRESSURES	VERIFY (SUPERSONIC) S&C/AIRLOADS MODEL
40-0	STRUCT. DYNAMICS	1.0	PANEL FLUTTER	TPS TILE / FLUTTER PANELS
41-0TS	THERMODYNAMICS	0.00593	THERMOCOUPLES	HEAT TRANSFER DISTRIBUTION MODEL

42-0	AERODYNAMICS	0.015	FORCE & MOMENTS	STABILITY & CONTROL / RCS MODEL
43-0	AERODYNAMICS	0.0405	FORCE & MOMENTS	VERIFICATION (SUBSONIC) STABILITY & CONTROL MODEL
44-0	AERODYNAMICS	0.015	FORCE & MOMENTS	STABILITY & CONTROL (REMOTE CONTROLLED) MODEL
45-0	AERODYNAMICS	0.03	FORCE & MOMENTS/PRESSURES	STABILITY & CONTROL ORBITER/CARRIER MODEL
46-0	THERMODYNAMICS	0.00593	PAINT	HEAT TRANSFER DISTRIBUTION MODEL
47-0TS	AERODYNAMICS	0.03	PRESSURES/FORCE	AIRLOADS MODEL
48-0	AERODYNAMICS	0.0125	FORCE & MOMENTS	STABILITY & CONTROL ORBITER/CARRIER SEP. MODEL
49-0	AERODYNAMICS	0.015	FORCE & MOMENTS	STABILITY & CONTROL MODEL
50-0	THERMODYNAMICS	0.006	THERMOCOUPLES	GAMMA EFFECT HEATING MODEL
51-0	AERODYNAMICS	0.01	FORCE & MOMENTS	STABILITY & CONTROL MODEL
52-0TS	AERODYNAMICS	0.01	FORCE & MOMENTS	STABILITY & CONTROL MODEL
53-0	THERMODYNAMICS	0.111	THERMOCOUPLES	FLAT PLATE: WING/ELEVON GAP SIMULATION
54-0	STRUCT. DYNAMICS	0.14	STRAIN GAGES	WING/ELEVON FLUTTER MODEL
55-0	STRUCT. DYNAMICS	0.14	STRAIN GAGES	VERTICAL/RUDDER FLUTTER MODEL
56-0TS	THERMODYNAMICS	0.0175	PAINT	PHASE CHANGE PAINT MODELS OF VERTICAL STABILIZERS
57-0	AERODYNAMICS	0.1	PRESSURES:AIR DATA PROBES	ORBITER FOREBODY (NOSE BOOM) MODEL
58-0	THERMODYNAMICS	1.0	THERMOCOUPLES & PRESS.	FLAT PLATE: SHOCK GENERATOR MODELS
59-0T	THERMODYNAMICS	0.01	THIN FILM GAGE / PRESS.	HEAT TRANSFER DISTRIBUTION MODEL
60-0TS	THERMODYNAMICS	0.0175	THERMOCOUPLES	THIN SKIN TEMPERATURE MODEL
61-0	THERMODYNAMICS	0.01	PRESSURES	AEROHEATING PRESSURE MODEL
62-0	-------CANCELLED, NOT BUILT			
63-0T	-------CANCELLED, NOT BUILT			
64-0	THERMODYNAMICS	0.0175	PAINT	WING SECTION (SHOCK GENERATOR) MODEL
65-0	THERMODYNAMICS	0.04	PRESSURES: BASE HEATING	BASE HEATING PLUME MODEL

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66-0	THERMODYNAMICS	0.025	THERMOCOUPLES	WING SECTION (SHOCK GENERATOR) THIN SKIN MOD.
67-0TS	AERODYNAMICS	0.015	FORCE & MOMENTS	STABILITY & CONTOL MODEL
68-T	AERODYNAMICS	0.07	FORCE & MOMENTS	STABILITY & CONTROL MODEL
69-0	AERODYNAMICS	0.015	FORCE & MOMENTS	STABILITY & CONTROL MODEL
70-0T	AERODYNAMICS	0.0125	FORCE & MOMENTS	STABILITY & CONTROL / RCS MODEL
71-0	-------CANCELLED, NOT BUILT			
72-0TS	AERODYNAMICS	0.01	FORCE & MOMENTS	STABILITY & CONTROL MODEL
73-0T	-------CANCELLED, NOT BUILT			
74-0TS	AERODYNAMICS	0.004	FORCE & MOMENTS	STABILITY & CONTROL MODEL
75-0TS	AERODYNAMICS	0.01	PRESSURES: JET EFFECTS	COLD-JET PLUME SIMULATION ASCENT MODEL
76-0	AERODYNAMICS	0.36	FORCE & MOMENTS/PRESSURES	STABILITY & CONTROL / PRESSURES MODEL
77-0	AERODYNAMICS	0.004	FORCE & MOMENTS	STABILITY & CONTROL (SPOILERS) MODEL
78-0	-------CANCELLED, NOT BUILT			
79-0TS	STRUCT. DYNAMICS	0.055	STRAIN GAGE/FLUTTER/RIGID	RIGID STABILITY CHECKOUT MODEL
80-0	STRUCT. DYNAMICS	0.055	STRAIN GAGE/FLUTTER	AEROELASTIC STRUCTURAL DYNAMIC MODEL
81-0	STRUCT. DYNAMICS	1.0	STRAIN GAGE:PANEL FLUTTER	TPS STRUCTURAL PANEL MODEL
82-0	THERMODYNAMICS	0.04	PAINT	HEAT TRANSFER FOREBODY MODEL
83-0	THERMODYNAMICS	0.04	THERMOCOUPLES	HEAT THIN-SKIN FOREBODY MODEL
84-0TS	STRUCT. DYNAMICS	0.035	PRESSURES	AIRLOADS/AERO NOIS MODEL
85-0	STRUCT DYNAMICS	1.0	THERMOCOUPLES:FRSI TILES	FLAT PLATE: FRSI TILE PANEL
86-0T	-------CANCELLED, NOT BUILT			
87-0TS	-------CANCELLED, NOT BUILT			
88-0TS	AERODYNAMICS	0.02	FORCE & PRESS: JET EFFECT	COLD-JET SIMULATION ASCENT MODEL
89-0	AERODYNAMICS	0.02	FORCE & MOMENTS	STABILITY & CONTROL MODEL

90-0	THERMODYNAMICS	0.005	THERMAL PAINT	HEAT TRANSFER DISTRIBUTION
91-0	THERMODYNAMICS	0.08	THERMOCOUPLES: WING TIPS	THIN-SKIN WING/WING TIP SEAL MODEL
92-0	THERMODYNAMICS	0.0175	PRESSURE	WING/ELEVON PRESSURE DISTRIBUTION MODEL
93-0	THERMODYNAMICS	0.04	THERMOCOUPLES	ELEVON/ELEVON SEAL MODEL
94-0	THERMODYNAMICS	0.03	PAINT	LOWER WING SURFACE SECTION MODEL
95-0	AERODYNAMICS	0.05	FORCE & MOMENTS	STABILITY & CONTROL MODEL
96-0	STRUCT. DYNAMICS	1.0	PRESSURES: LRSI TILE	FLAT PLATE: LRSI TILE PANEL
97-0	AERODYNAMICS	0.03	FORCE & MOMENTS	STABILITY & CONTROL MODEL
98-0	STRUCT. DYNAMICS	1.0	PRESSURE: HRSI TILE	FLAT PLATE: HRSI TILE PANEL
99-0	AERODYNAMICS	0.1	PRESSURES: AIR DATA PROBES	ORBITER FOREBODY MODEL
100-OTS	STRUCT. DYNAMICS	0.046	DYNAMICS: GROUND WINDS	LAUNCH/LAUNCH PAD SIMULATION MODEL
101-0	-------CANCELLED, NOT BUILT			
102-0	-------CANCELLED, NOT BUILT			
103-0	-------CANCELLED, NOT BUILT			
104-0	AERODYNAMICS	0.02	FORCE & MOMENTS	VERIFICATION TRANSONIC HM MODEL (MOD. #89)
105-0	AERODYNAMICS	0.02	FORCE & MOMENTS	VERIFICATION S&C MODEL
106-0	AERODYNAMICS	0.02	FORCE & MOMENTS	VERIFICATION S&C R/C MODEL (MOD. #105)
107-0	AERODYNAMICS	1.0	PRESSURES	TPS TILE CAVITY FLOW FIELD MODEL
108-0	STRUCT. DYNAMICS	1.0	PRESSURES	ORBITER/ET TPS CAVITY MODEL
109-0	STRUCT. DYNAMICS	1.0	PRESSURES	ORBITER/ET DOOR TPS MODEL
110-0	STRUCT. DYNAMICS	1.0	PRESSURES	ELEVON/ELEVON GAP MODEL
111-0	STRUCT. DYNAMICS	1.0	PRESSURES	VERTICAL/RUDDER GAP MODEL
112-T	STRUCT. DYNAMICS	0.25	PRESSURE	EXTERNAL TANK FOREBODY PROTUBERANCE MODEL
113-0	STRUCT. DYNAMICS	1.0	PRESSURES	VENT PORT MODEL

10. DOCUMENTATION

Recognition of the need for an integrated, standardized system for processing, storing, and manipulating large blocks of wind tunnel data led to the development of the "System for Automated Development of Static Aerothermodynamic Criteria" in 1966 by Chrysler Corporation Space Division, New Orleans, Louisiana. Typically, raw wind tunnel data counts were automatically reduced to coefficient form at the facility, with many of the subsequent operations done predominantly by hand. Data point corrections, bias shifts, adjustments to the data for scale effects, breakdown of the data for component analysis, plots (and cross plots) for data evaluation, faired and interpolated data, final presentation plots, etc. were all done manually. Structural design related tests such as loads and heating were handled in a similar manner. However, there are usually many more data points for each test condition. In 1966, Chrysler, in support of the Saturn IB/Apollo Program, and under contract to the MSFC, designed and developed a digital computer program system which would include data file storage and retrieval operations, data computational capability, and automated plotting capability. In 1970, the original system, renamed DATAMAN (short for data management), was proposed by the MSFC as a means to document and file experimental wind tunnel data from the SSV design and development program. This proposal was accepted and the system became operational during the Phase B portion of the Space Shuttle Program. Since the beginning of Phase C/D, approximately one thousand test reports (of which 35 percent were special requests) have been issued in support of the SSV program. Each of these documents is retrievable and referenceable (abstracted in the Scientific and Technical Aerospace Reports, or STAR) and contains complete test information. However the test data is primarily in standard plots and tables, with limited analysis. This method of documenting test data is very unique. The current wind tunnel investigation for a given design results in generating, not only a large amount of data, but the data is obtained from a number of different facilities and is often provided in different nomenclature and format. The impact on the engineering analyst, with limited resources available to organize, manipulate, and plot data, is to either delay release of the data or to limit the scope of the analyses. By automating and standardizing these procedures, as was done for the SSV program, rapid output of the data in the desired reference system and format maximizes its use by the engineering community. These techniques have proven to be both an effective and economical method of documenting, as well as working with, wind tunnel test results and could be applied to any major aerospace program.

A summary of the many advantages of this system follows. The system enables the engineer to spend more time evaluating and analzying the data by relieving the tedious job of organizing and plotting large quantities of data as a necessary prelude to analysis and evaluation. The initial output of plotted data is available in a time frame that permits rapid evaluation. Thus the engineer can incorporate any findings into subsequent wind tunnel investigations. The system permits maximum exploitation of the data by allowing a complete analysis of major and second order effects by providing the engineer with plots of all of the data, which would not normally be possible with limited manpower. The method permits extensive analysis of the data by providing automated calculation and comparative plotting of such variables as intercepts, derivatives, increments, trim conditions, control power effectiveness, etc. These are operations which must be performed for design applications of the data and consequently are equally important to the engineer. The system facilitates comparison of data from current tests with data from prior tests by providing a data storage and retrieval system. It expedites the publication and dissemination of all the data obtained. The system provides a document containing a complete set of the data obtained in the investigation. These documents can subsequently be referenced in more formal documents and presentations. A complete listing of all test documents produced is given in appendix B. Table B1 relates the document number to the test number, test data, model reference configuration and ID, and the facility. Table B2 shows the NASA contractor report (CR) number, the test number, and the report title.

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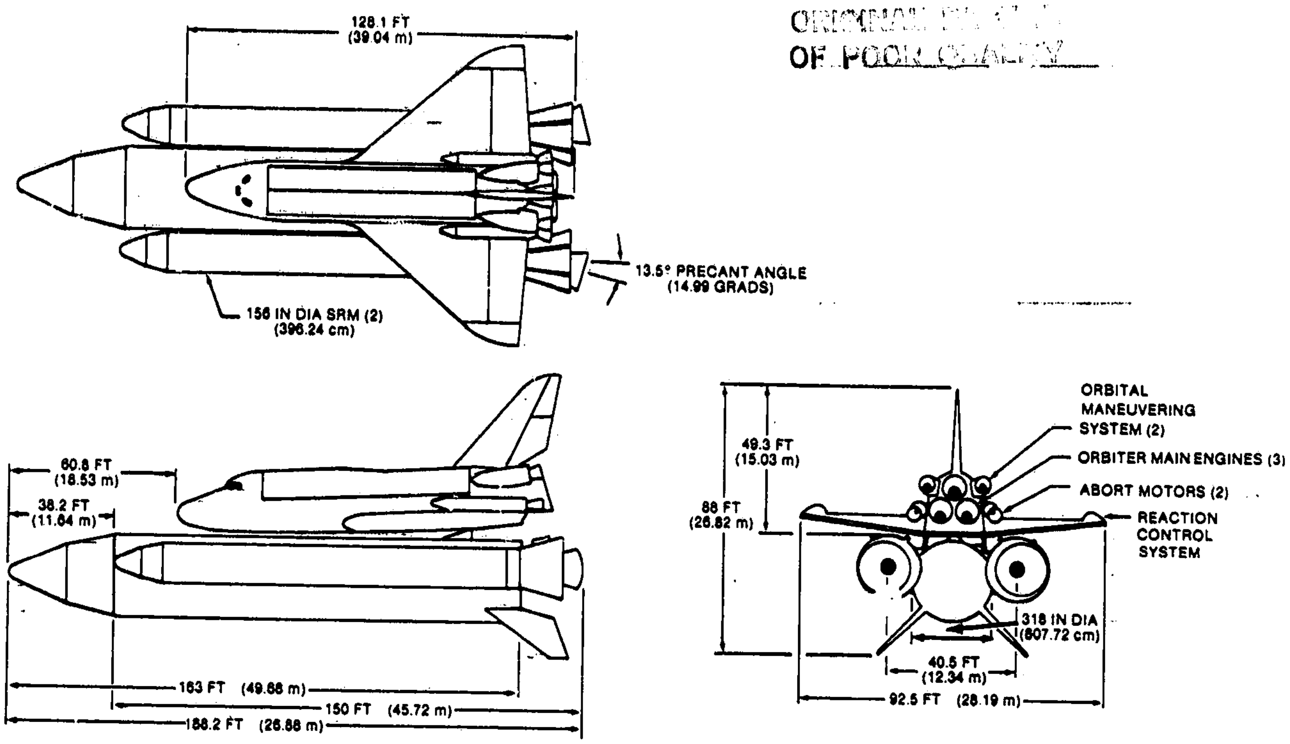
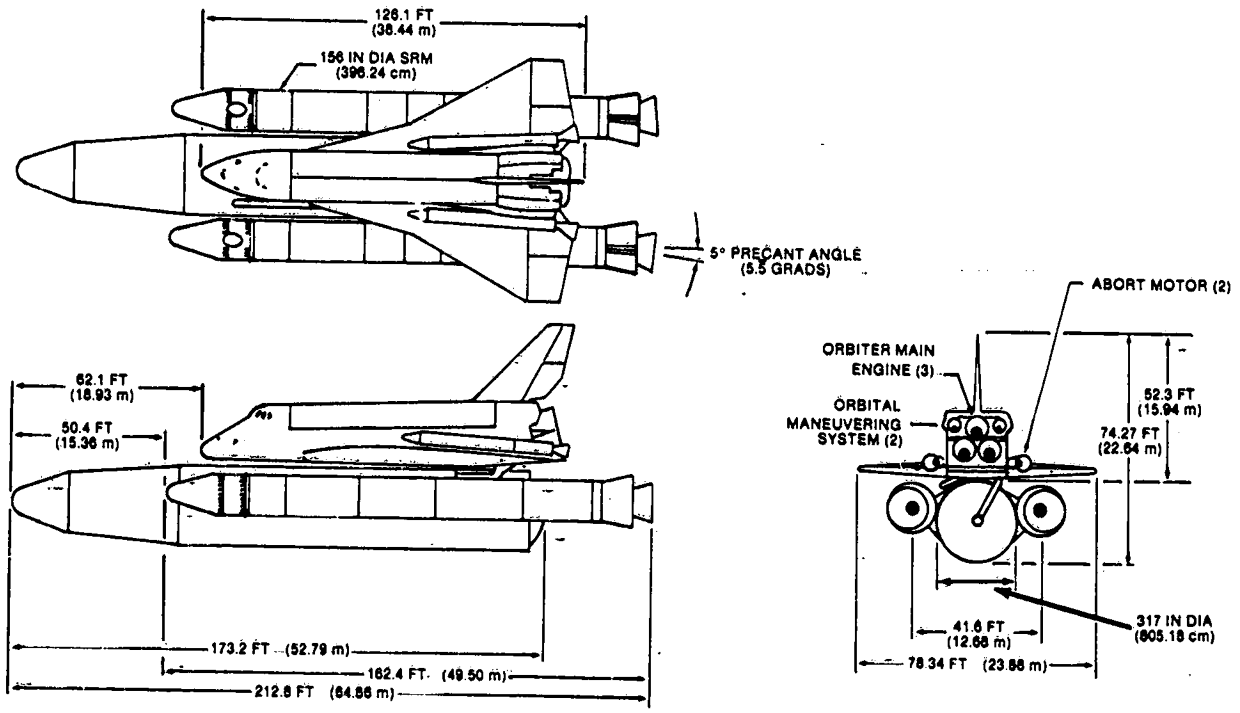


(a) GRUMMAN AIRCRAFT COMPANY CONFIGURATION



(b) LOCKHEED MISSILE AND SPACE COMPANY

Figure 2.1. - Space Shuttle Phase B Double Prime final configurations.

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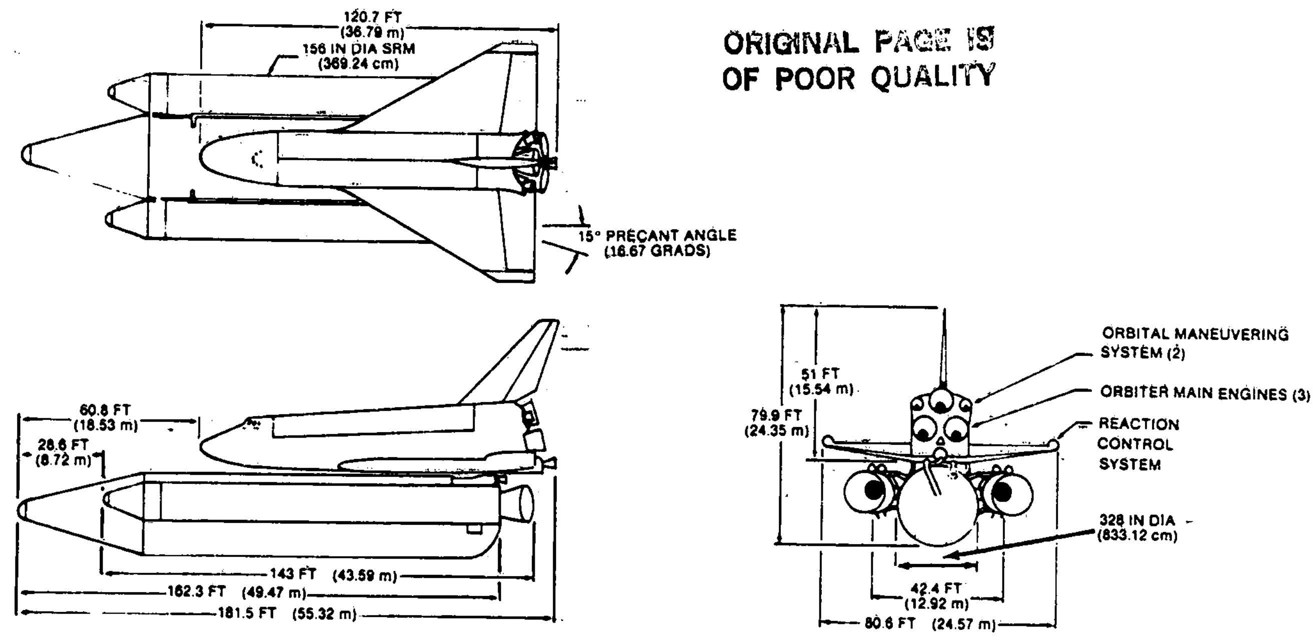
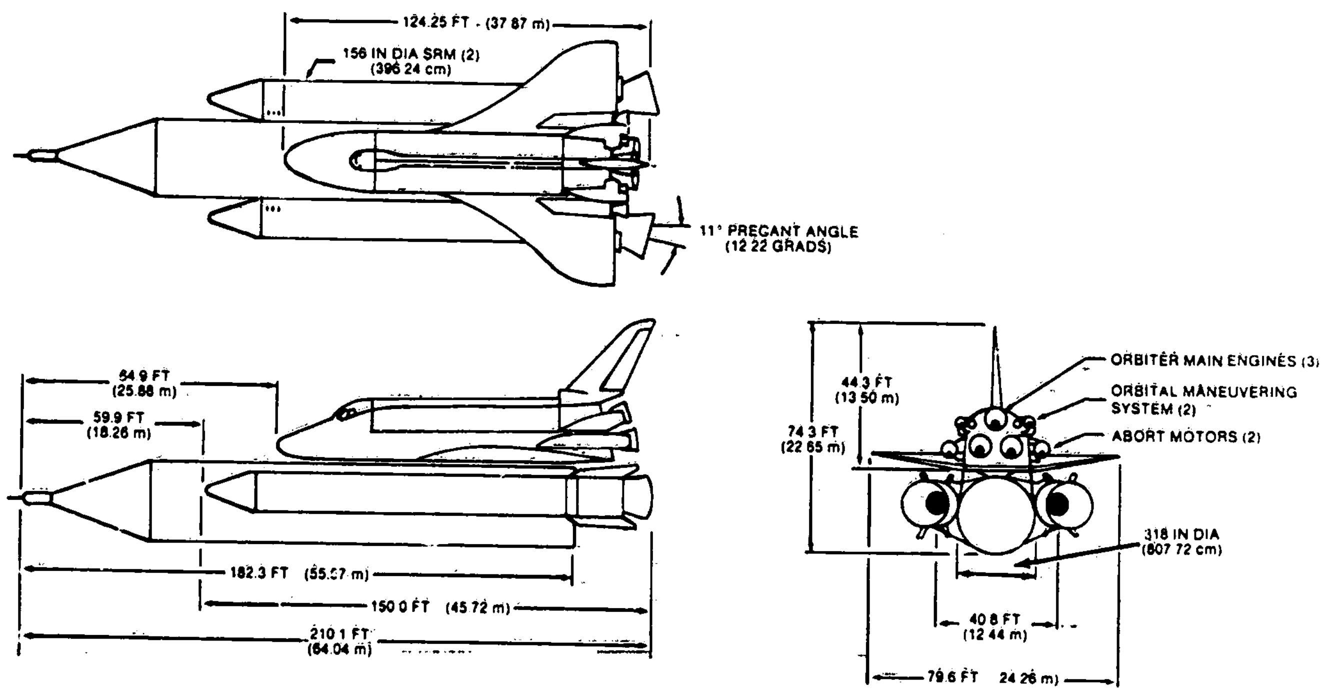


(c) McDONNELL DOUGLAS CONFIGURATION



(d) NORTH AMERICAN/ROCKWELL CONFIGURATION

Figure 2.1. - Concluded

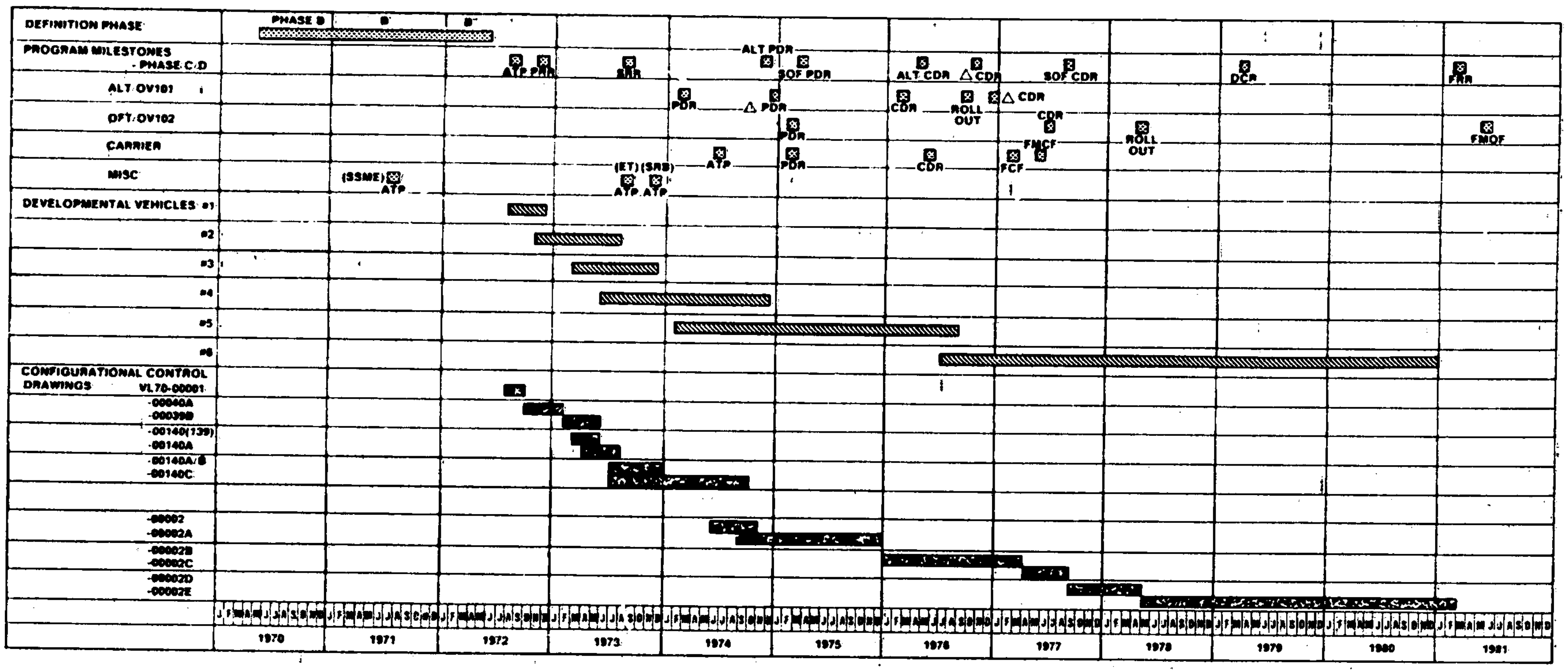
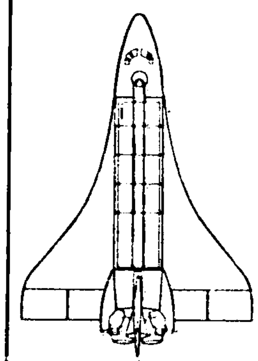
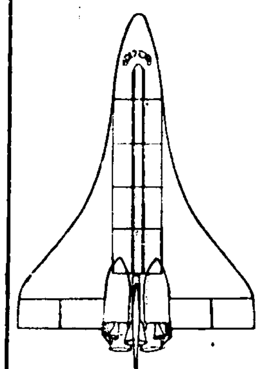
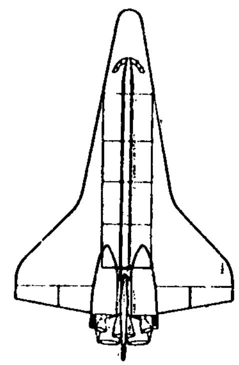
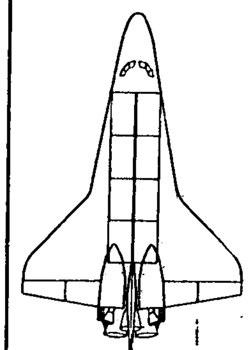
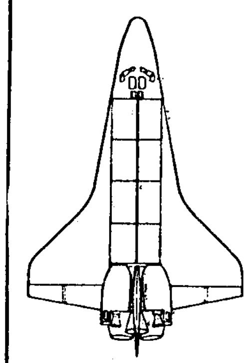
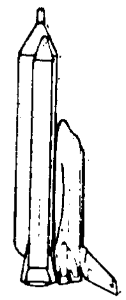
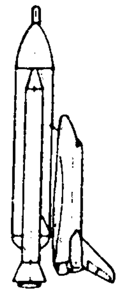
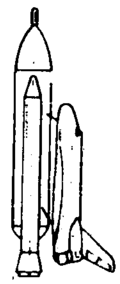
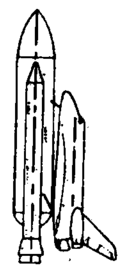
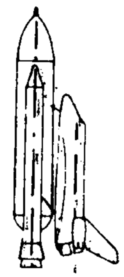


Figure 3.1. - Space Shuttle configuration evolution timeline.

CONFIGURATION DESIGNATION	ATP	PRR	2A	3,4	5,6
CONFIGURATION CONTROL DRAWING NUMBER	VL70 - 000001	VL70 - 000040A	VL70 - 000089B	VL70 - 000140, 140A/B	VL70 - 000140C, VC70 - 000002
WING DESIGN	50° BLENDED DELTA	50° BLENDED DATA	45°/79° DOUBLE DELTA	45°/81° DOUBLE DELTA	45°/81° DOUBLE DELTA
WING AREA, FT^2 (m^2)	3220 (299.14)	3220 (299.14)	2690 (249.90)	2690 (249.90)	2690 (249.90)
WING SPAN, FT (m)	84.0 (25.60)	84.0 (25.60)	78.1 (23.80)	78.1 (23.80)	78.1 (23.80)
OVERALL LENGTH, FT (m)	125.8 (38.34)	125.8 (38.34)	125.2 (38.16)	122.8 (37.43)	122.2 (37.25)
PLAN VIEW					
DRY WEIGHT, LB (kg)	170,000 (77,110)	170,000 (77,110)	150,000 (68,039)	150,000 (68,039)	150,000 (68,039)
LANDING PAYLOAD, LB (kg)	40,000 (18,144)	40,000 (18,144)	25,000 (11,340)	32,000 (14,515)	32,000 (14,515)
CG RANGE (% REFERENCE LENGTH)	65.0 - 68.0	65.0 - 68.0	66.0 - 68.0	65.0 - 67.5	65.0 - 67.5

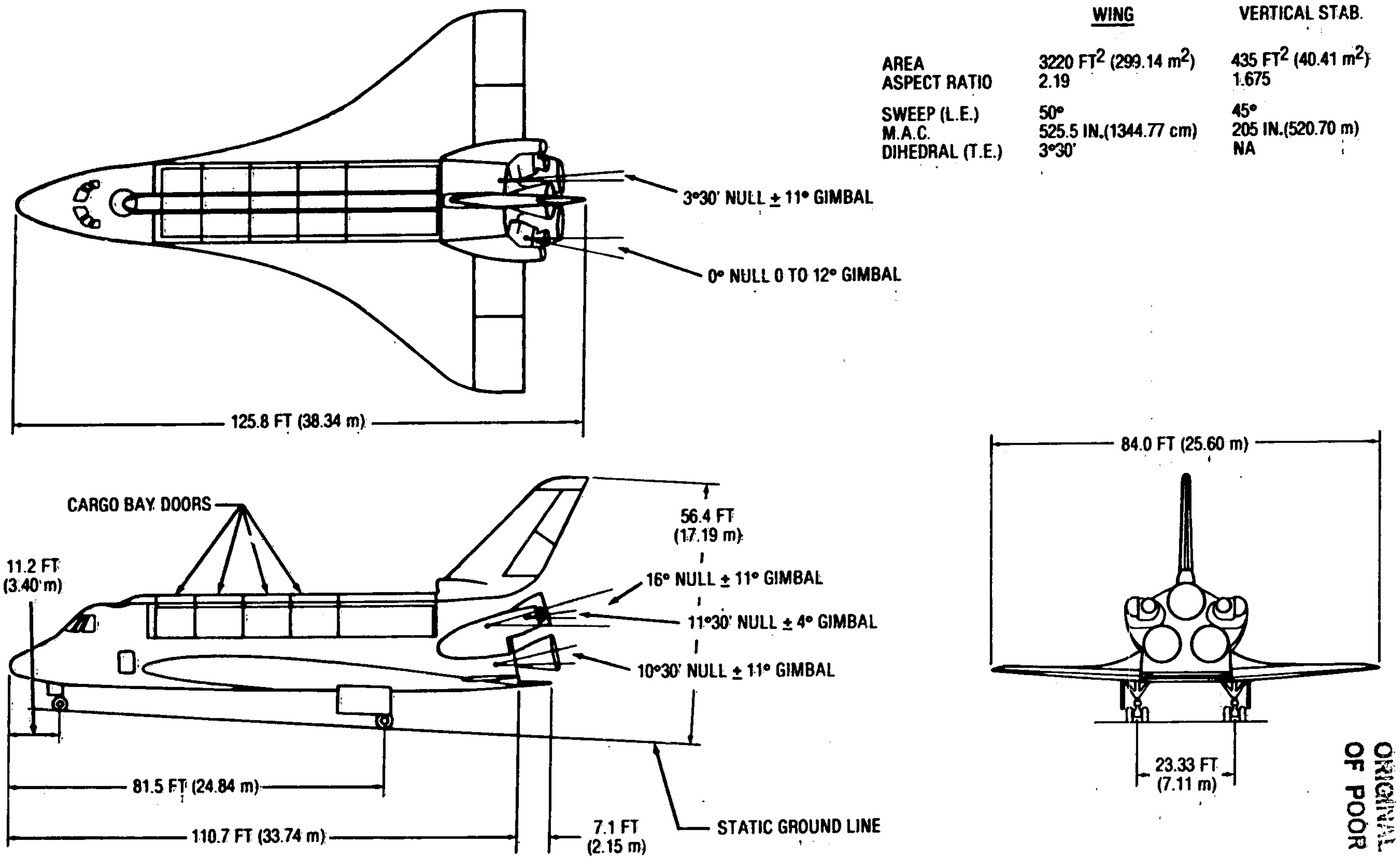
(a) Orbiter Vehicle.

Figure 3.2. - Major configuration evolution definitions.

CONFIGURATION DESIGNATION	ATP	PRR	VEH. 2A	VEH. 3,4	VEH. 5,6
CONFIGURATION CONTROL DRAWING NUMBER	VL72 - 000001	VL72 - 000030	VL72 - 000061A	VL72 - 000088A, 88B	VL72 - 000143D, VC72 - 000002C
OVERALL LENGTH, FT (m)	205.7 (62.70)	214.3 (65.32)	192.3 (58.61)	181.3 (55.26)	183.6 (55.96)
ET LENGTH, FT (m)	182.0 (55.47)	189.8 (57.85)	165.8 (50.54)	155.4 (47.37)	153.7 (46.85)
ET DIAMETER, IN (cm)	318.0 (807.7)	304 (772.2)	324 (823.0)	324 (823.0)	331.0 (840.7)
ET NOSE SHAPE	30° BLUNTED CONE	OGIVE (568" R)	OGIVE (605" R)	OGIVE (600" R)	OGIVE (612" R)
ET NOSE TIP LENGTH, FT (m)	10.[illegible] (3.15)	10.33 (3.15)	11.58 (3.53)	NONE	1.57 (0.48) SPIKE
SRB LENGTH, FT (m)	184.8 (56.33)	175.1 (53.37)	145.1 (44.23)	145.1 (44.23)	149.1 (45.45)
SRB DIAMETER, IN (cm)	156 (396.2)	162 (411.5)	142.3 (361.4)	142.3 (361.4)	146.0 (370.8)
SRB DISTANCE AFT OF ET TIP, FT (m)	17.5 (5.33)	39.3 (11.98)	47.3 (14.42)	36.2 (11.03)	34.5 (10.52)
ORBITER DISTANCE AFT OF ET TIP, FT (m)	80.3 (24.48)	88.6 (27.01)	80.8 (24.63)	56.7 (17.28)	53.9 (16.43)
PROFILE VIEW					
SRB GIMBAL SETTING, DEGREES	11° YAW FIXED	3.5° YAW ± 5°	0° ± 5°	0° ± 5°	0° ± 8°
ET PROPELLANT, THOUSAND POUNDS (THOUSAND kg)	1697 (769.7)	1650 (748.4)	1550 (703.1)	1549 (702.6)	1552 (704.0)
BOOSTER LIFT OFF WEIGHT, THOUSAND POUNDS (THOUSAND kg)	3252 (1475.1)	3276 (1486.0)	2259 (1024.7)	2327 (1055.5)	2327 (1055.5)
GROSS LIFT OFF WEIGHT, THOUSAND POUNDS (THOUSAND kg)	5411 (2454.4)	5261 (2386.3)	4116 (1867.0)	4183 (189[illegible])	4197 (1903.7)

(b) Integrated Vehicle.

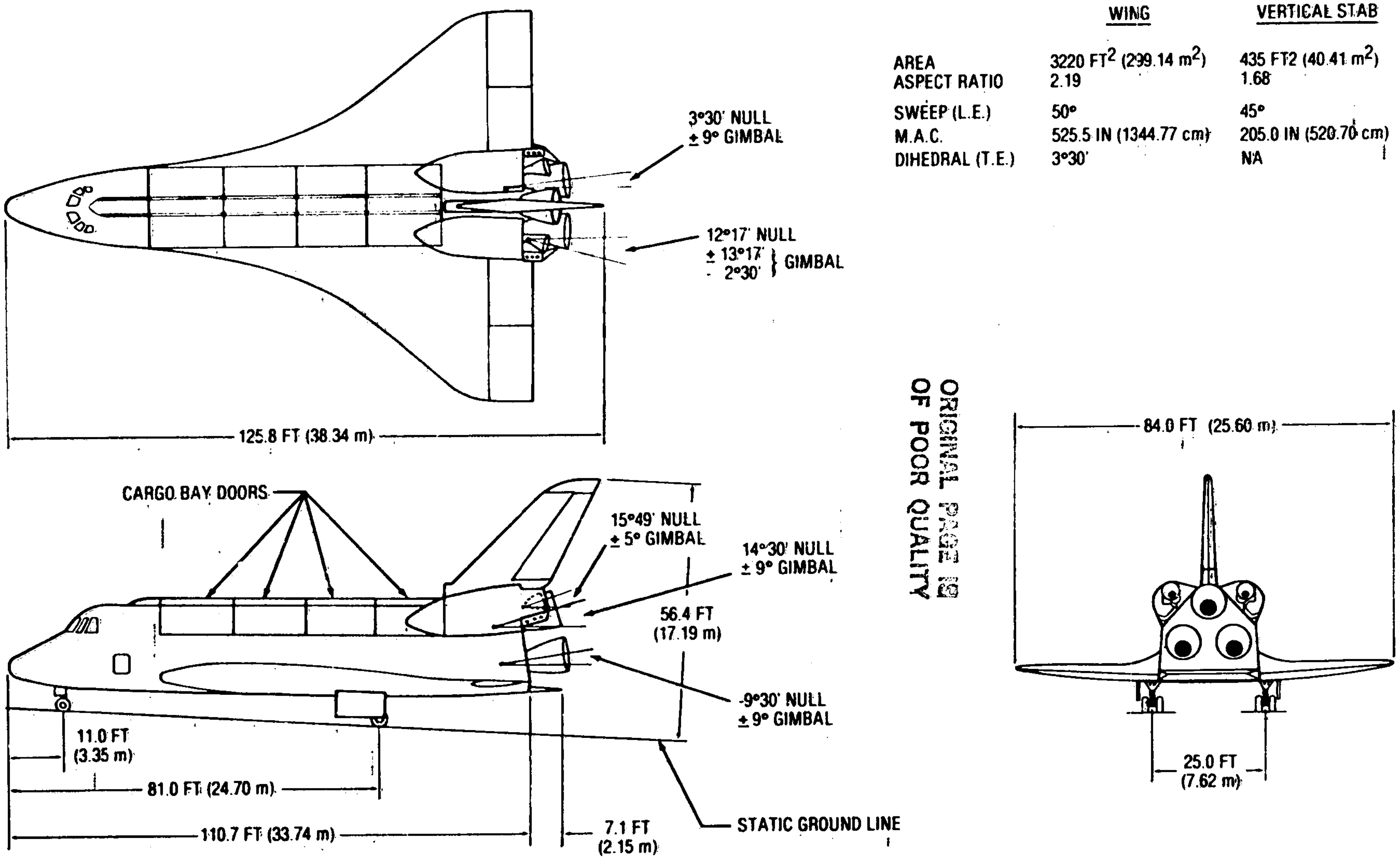
Figure 3.2. - Concluded.



	WING	VERTICAL STAB.
AREA	3220 FT^2 (299.14 m^2)	435 FT^2 (40.41 m^2)
ASPECT RATIO	2.19	1.675
SWEEP (L.E.)	50°	45°
M.A.C.	525.5 IN.(1344.77 cm)	205 IN.(520.70 m)
DIHEDRAL (T.E.)	3°30'	NA

(a) ATP configuration.

Figure 3.3. - Orbiter Vehicle dimensions; configuration evolution.

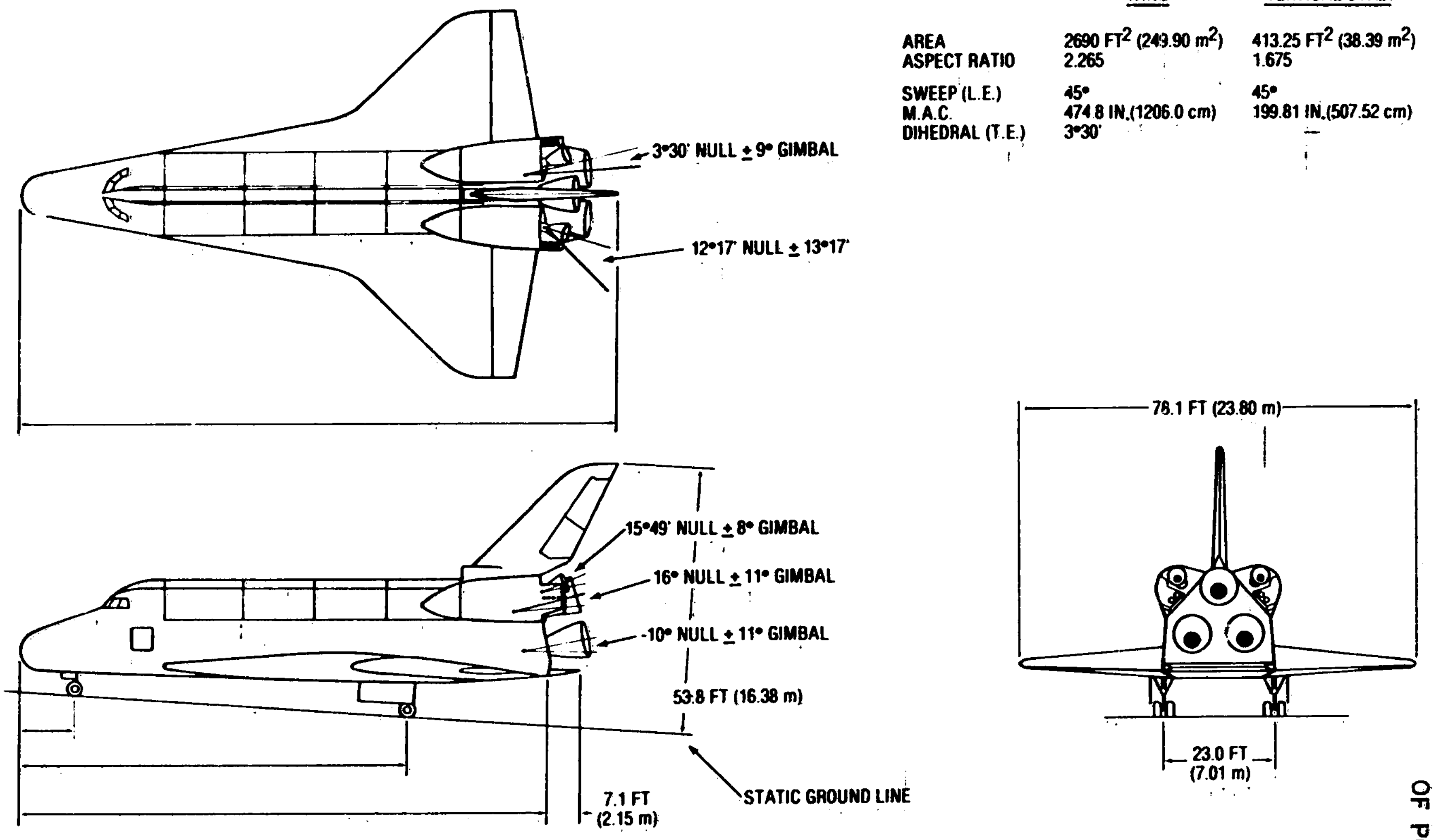


	WING	VERTICAL STAB
AREA	3220 FT^2 (299.14 m^2)	435 FT2 (40.41 m^2)
ASPECT RATIO	2.19	1.68
SWEEP (L.E.)	50°	45°
M.A.C.	525.5 IN (1344.77 cm)	205.0 IN (520.70 cm)
DIHEDRAL (T.E.)	3°30'	NA

(b) PRR configuration.

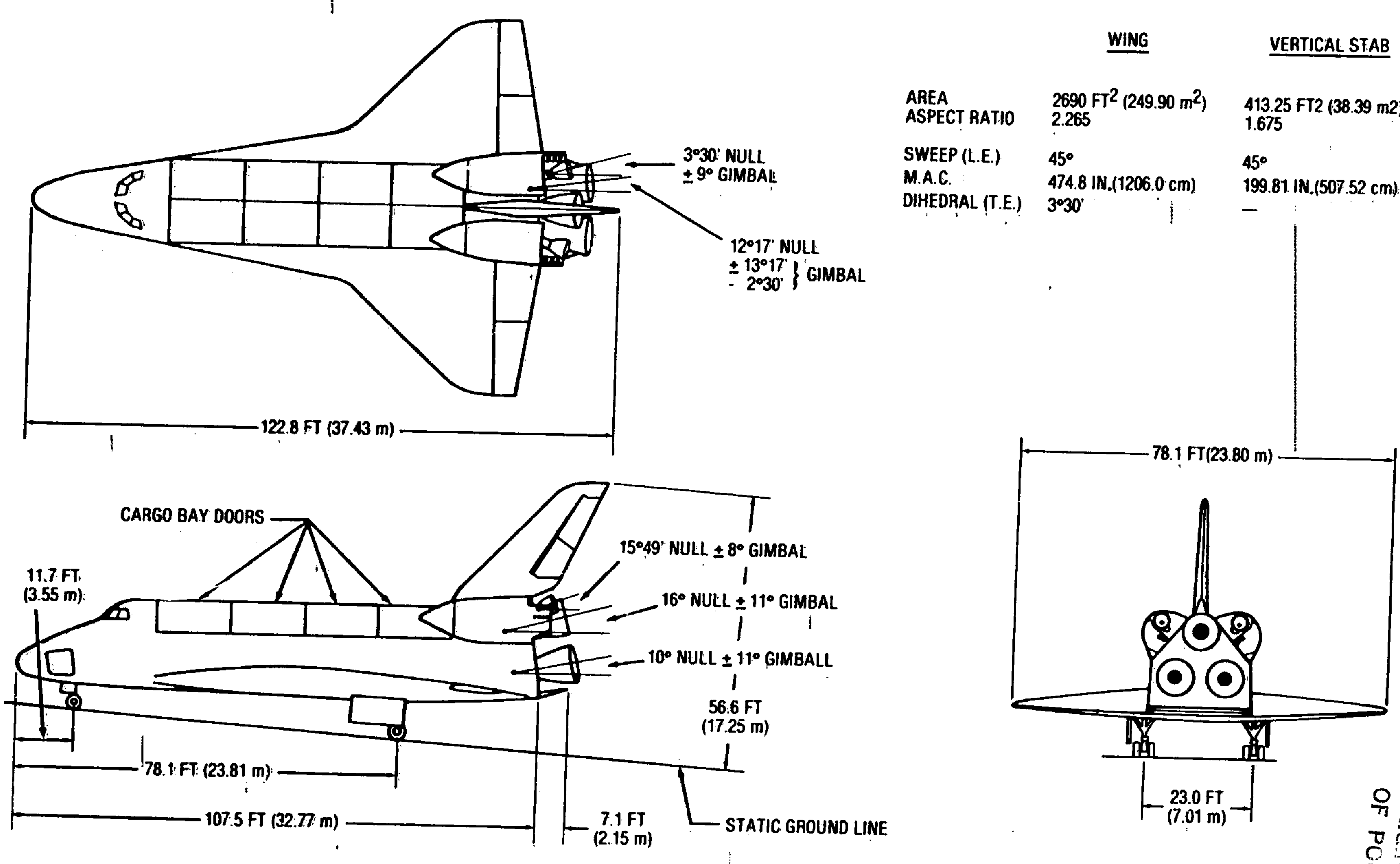
Figure 3.3. - Continued.

	WING	VERTICAL STAB.
AREA	2690 FT^2 (249.90 m^2)	413.25 FT^2 (38.39 m^2)
ASPECT RATIO	2.265	1.675
SWEEP (L.E.)	45°	45°
M.A.C.	474.8 IN.(1206.0 cm)	199.81 IN.(507.52 cm)
DIHEDRAL (T.E.)	3°30'	



(c) Vehicle 2A.

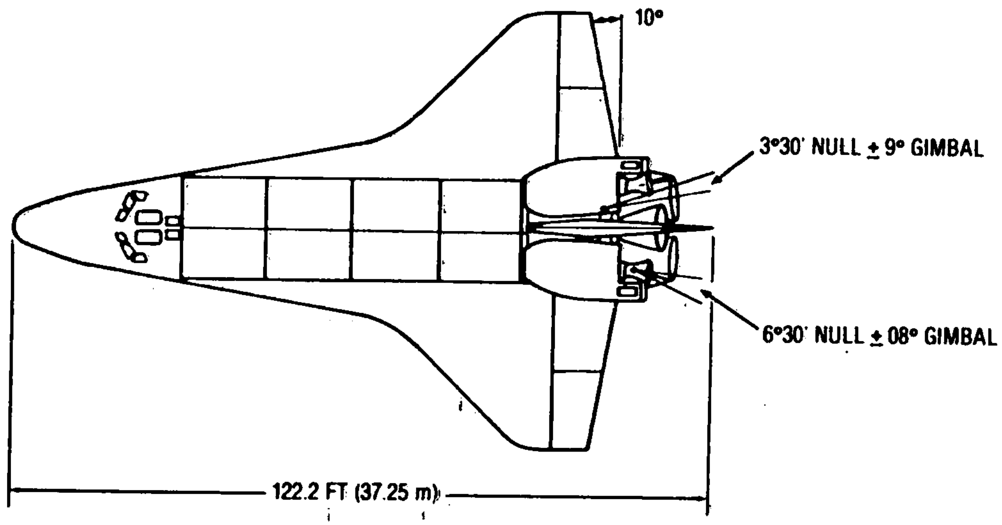
Figure 3.3. - Continued.



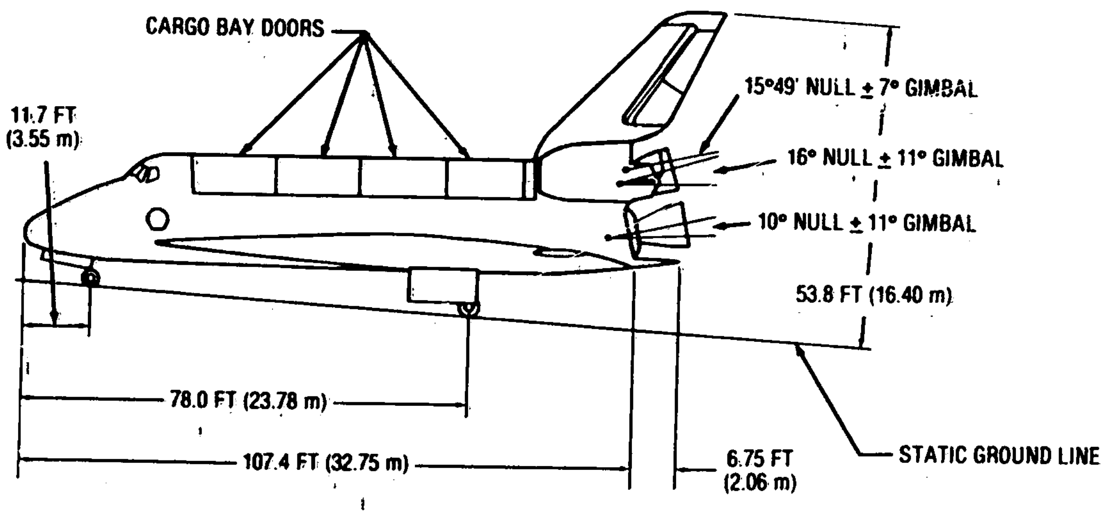
	WING	VERTICAL STAB
AREA	2690 FT^2 (249.90 m^2)	413.25 FT2 (38.39 m2)
ASPECT RATIO	2.265	1.675
SWEEP (L.E.)	45°	45°
M.A.C.	474.8 IN.(1206.0 cm)	199.81 IN.(507.52 cm)
DIHEDRAL (T.E.)	3°30'	—

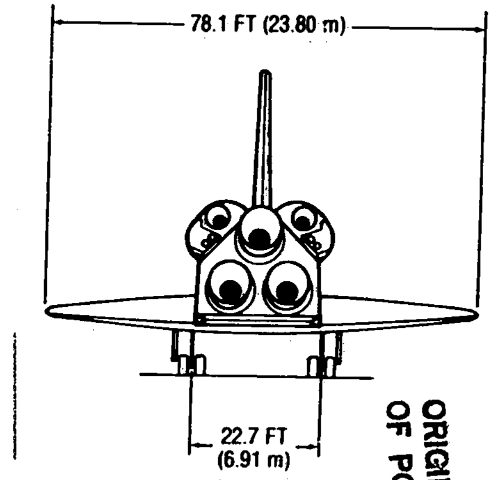
(d) Vehicles 3, 4.

Figure 3.3. - Continued.



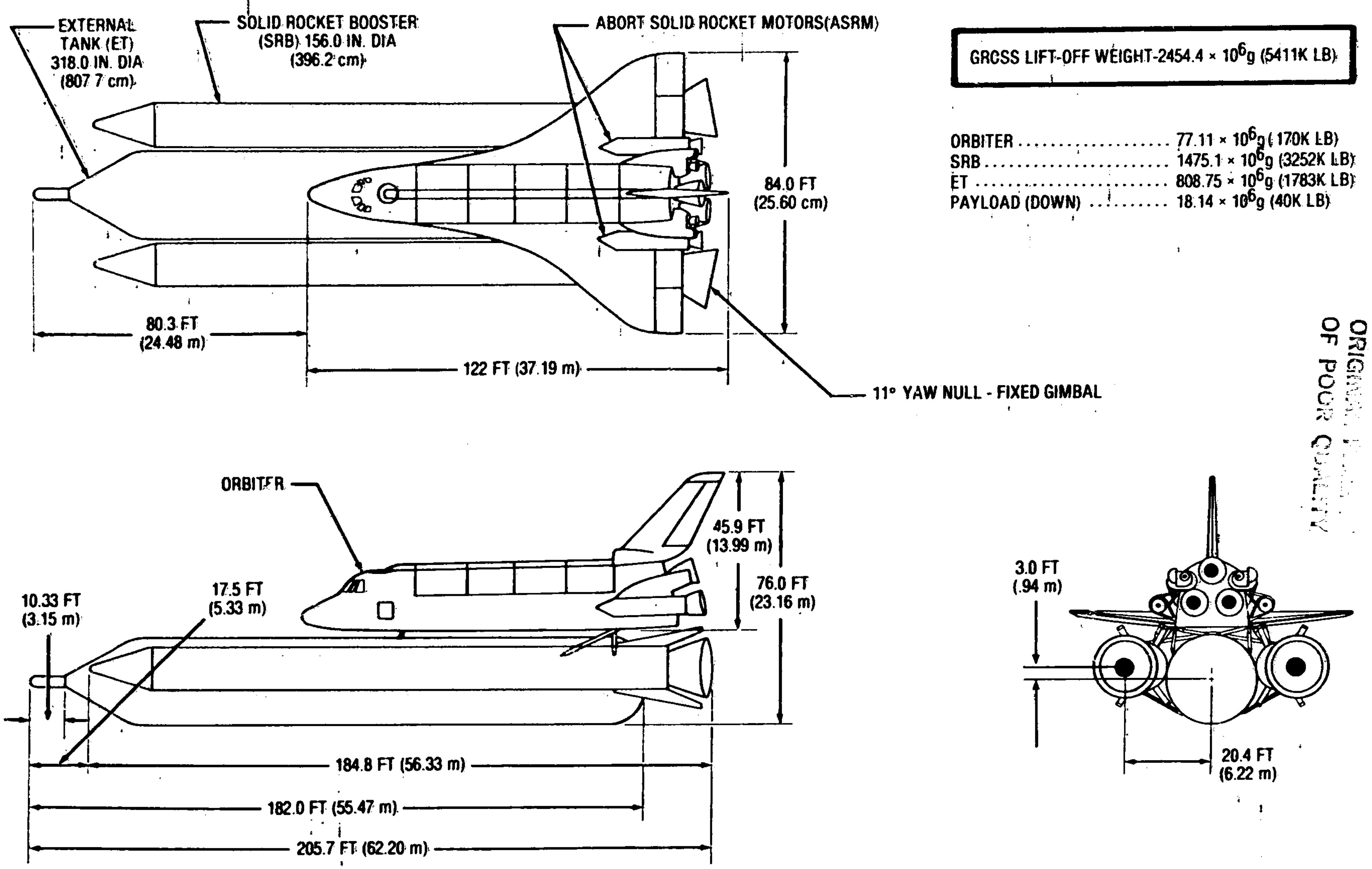
	WING	VERTICAL STAB.
AREA	2690 FT^2 (249.90 m^2)	413.25 FT^2 (38.39 m^2)
ASPECT RATIO	2.265	1.675
SWEEP (L.E.)	45°	45°
M.A.C.	474.81 IN.(1206.0 cm)	199.81 IN.(507.52 cm)
DIHEDRAL (T.E.)	3°30'	—





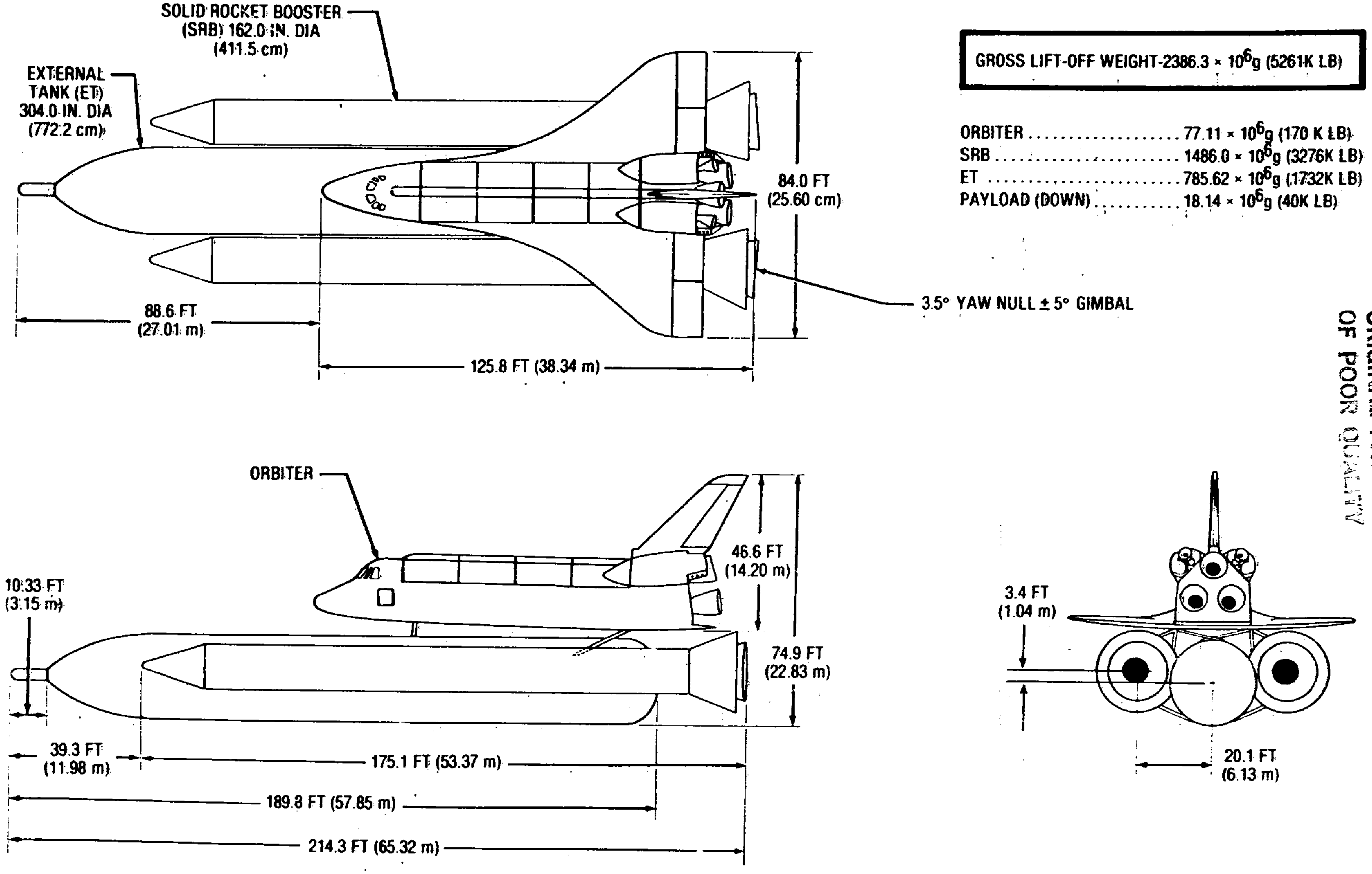
(e) Vehicles 5, 6.

Figure 3.3. - Concluded.



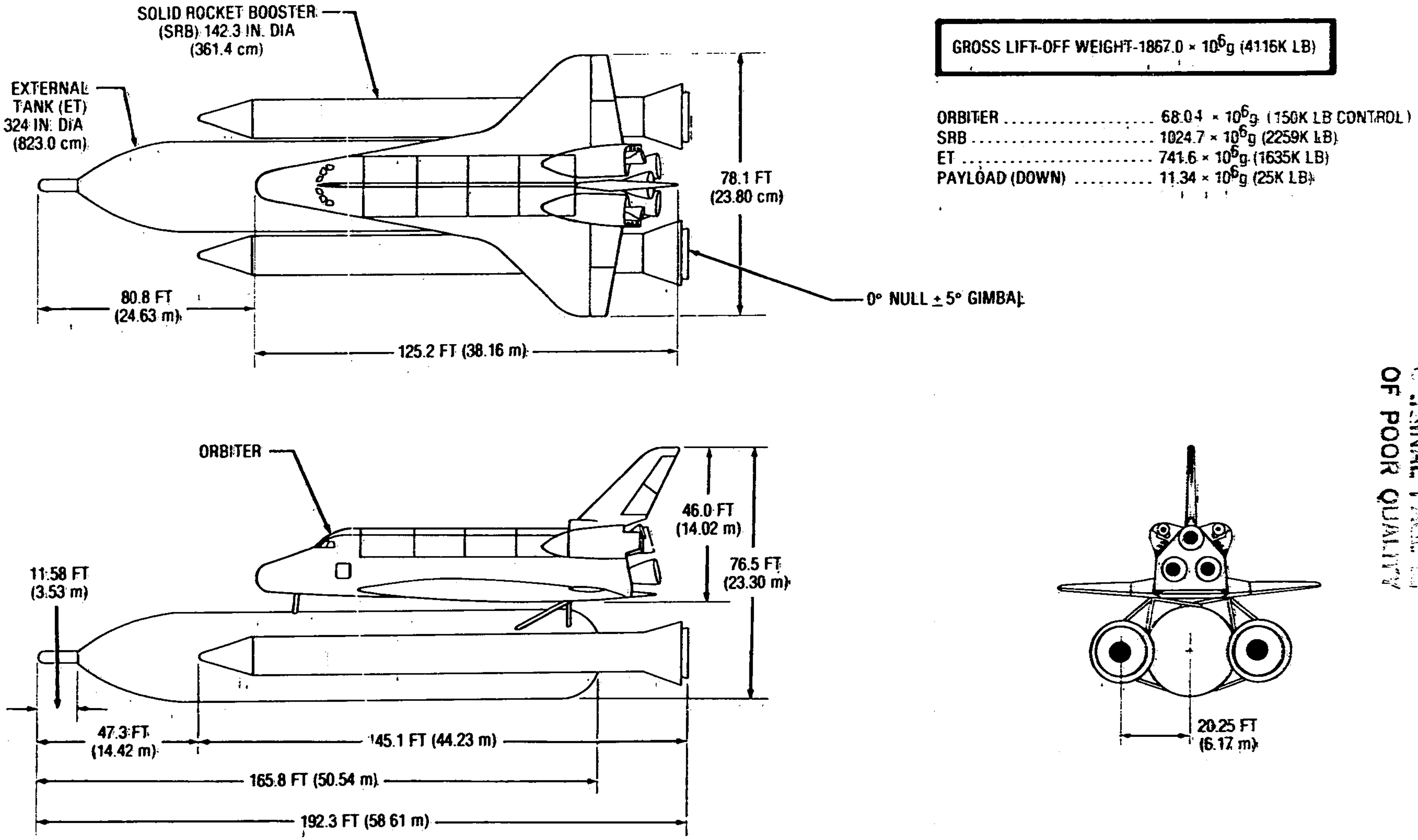
(a) ATP configuration.

Figure 3.4. - Integrated Vehicle dimensions; configuration evolution.



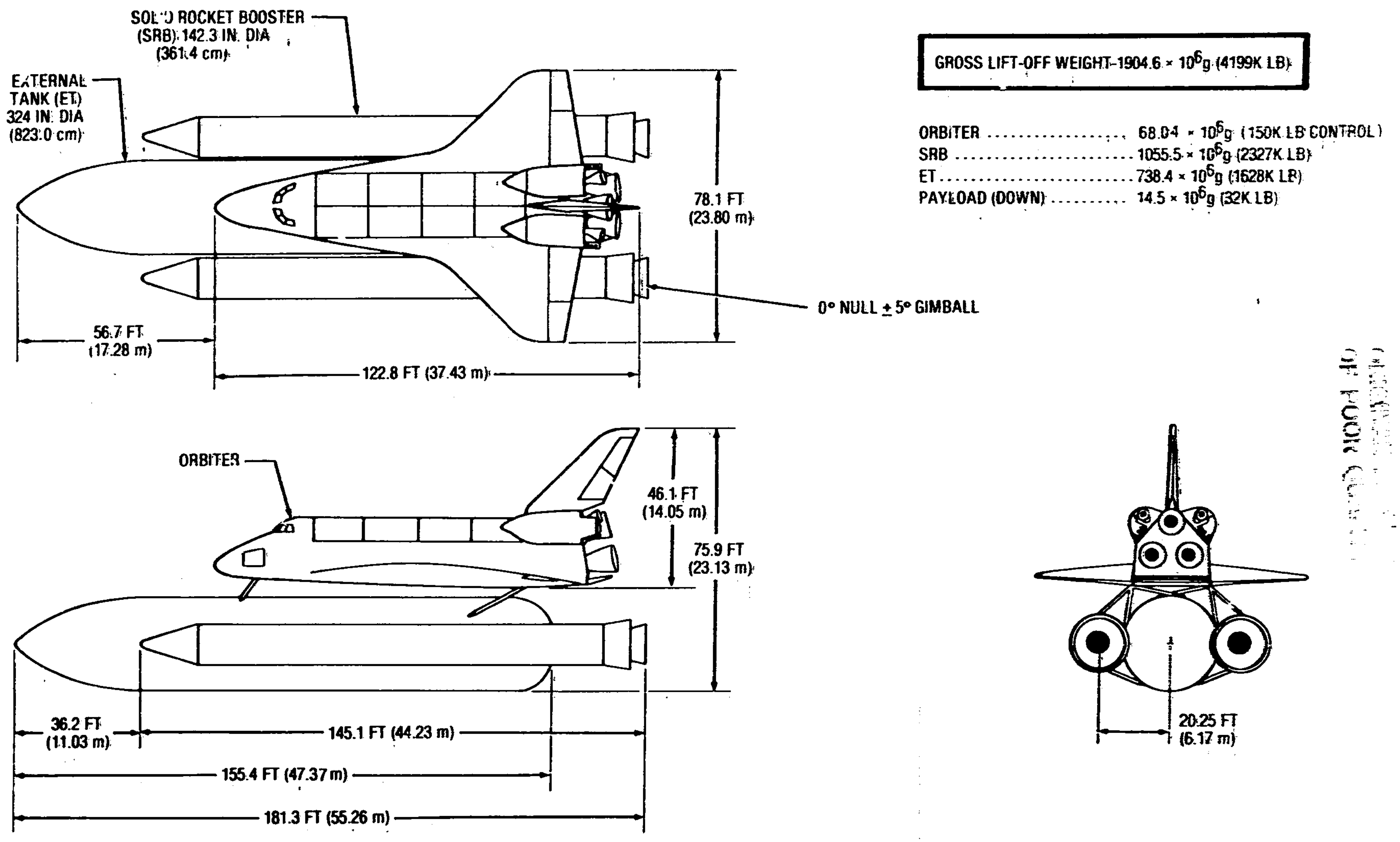
(b) PRR configuration.

Figure 3.4. - Continued.



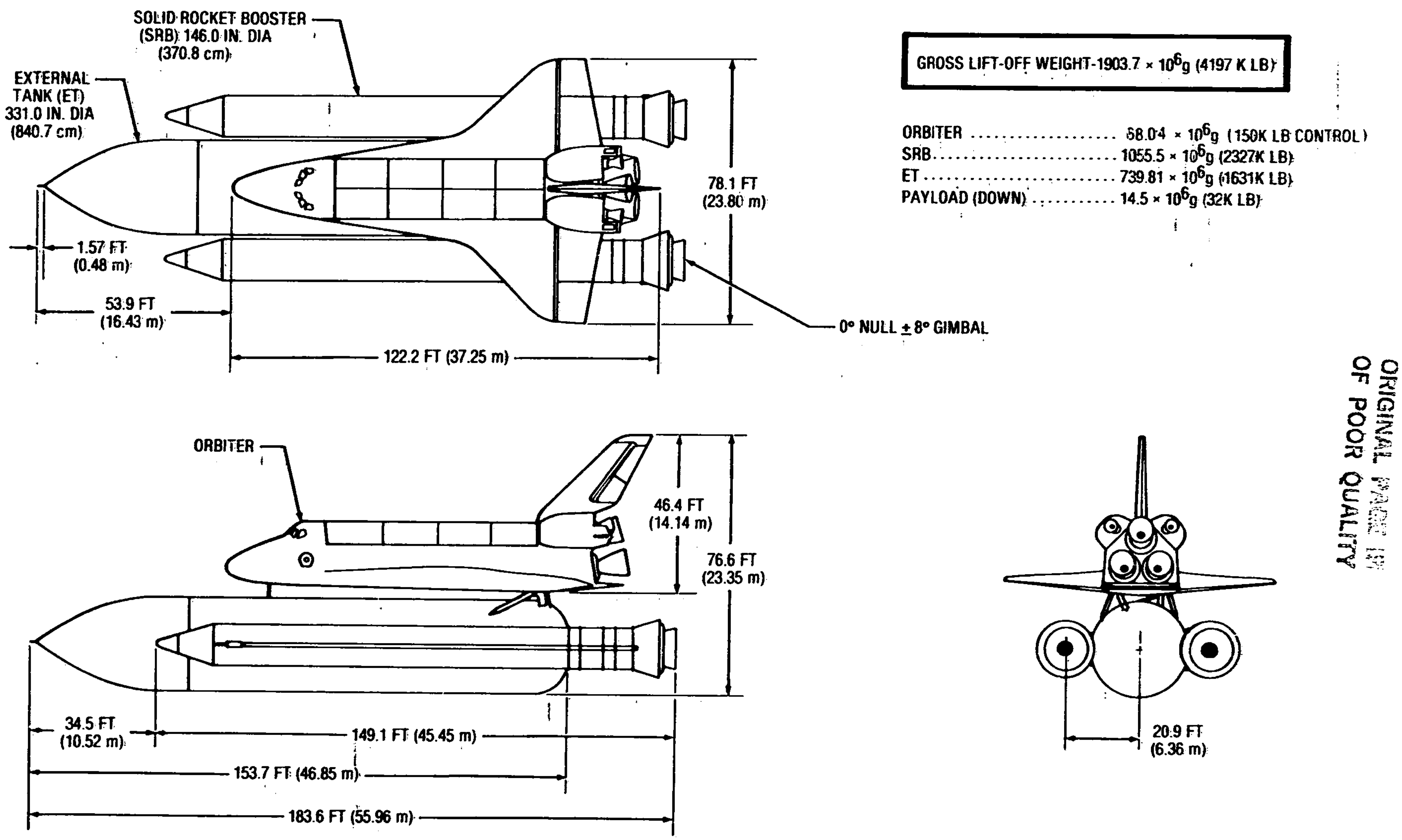
(c) Vehicle 2A.

Figure 3.4. - Continued.



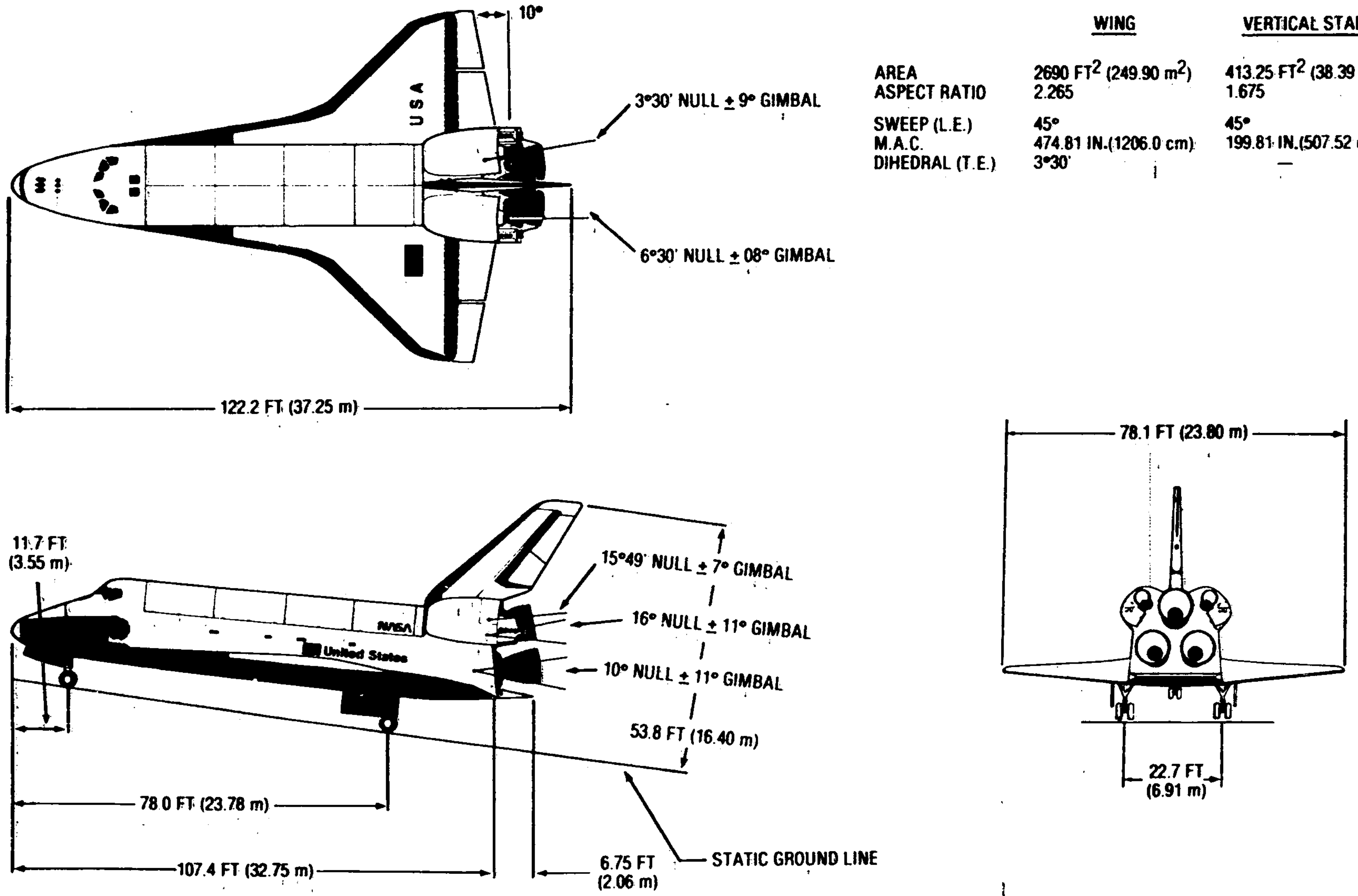
(d) Vehicles 3, 4.

Figure 3.4. - Continued.



(e) Vehicles 5, 6.

Figure 3.4. - Concluded.

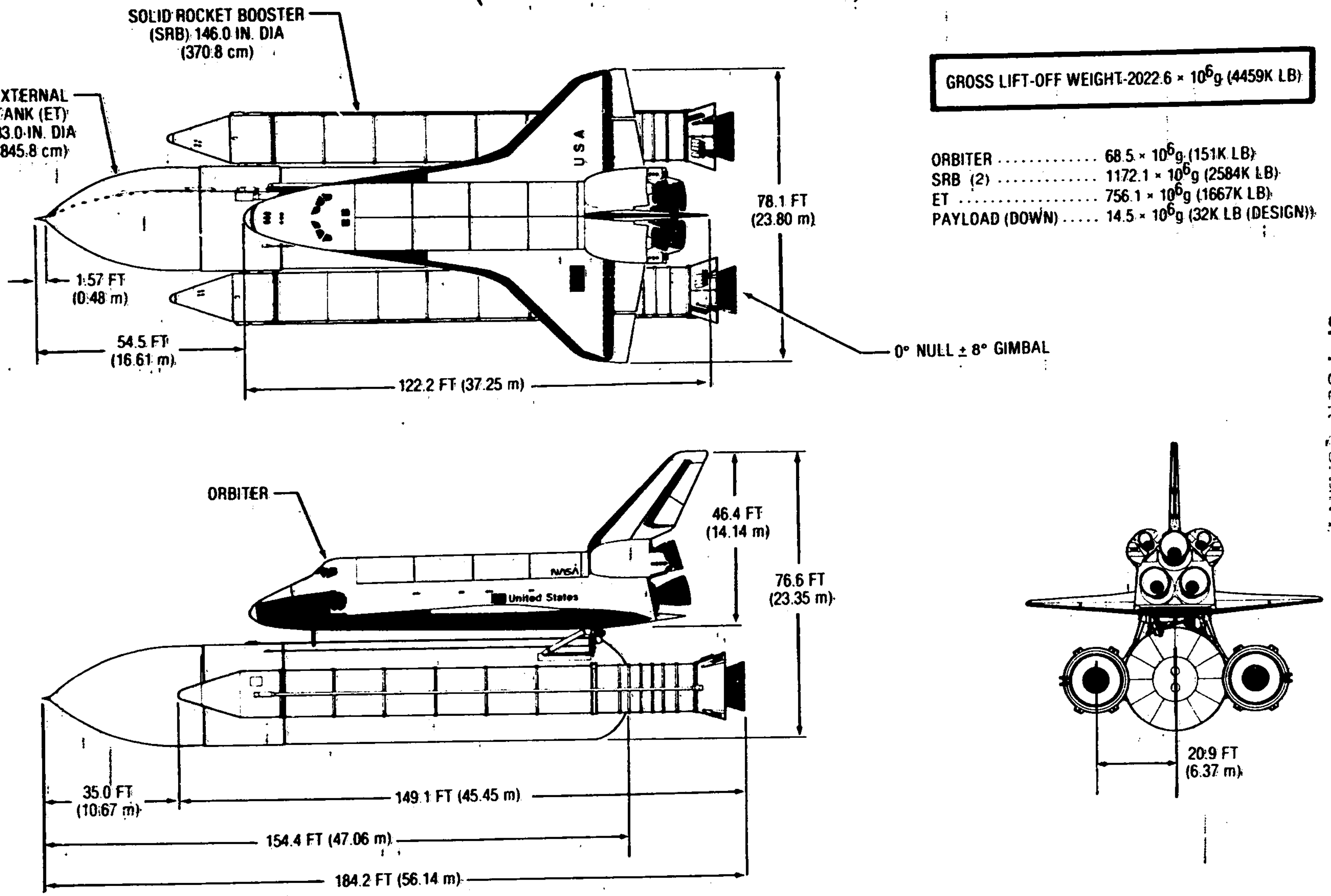


	WING	VERTICAL STAB.
AREA	2690 FT^2 (249.90 m^2)	413.25 FT^2 (38.39 m^2)
ASPECT RATIO	2.265	1.675
SWEEP (L.E.)	45°	45°
M.A.C.	474.81 IN.(1206.0 cm)	199.81 IN.(507.52 cm)
DIHEDRAL (T.E.)	3°30'	—

(a) Orbiter Vehicle OV-102.

Figure 3.5. - STS-1 mission configurations.

SPACE SHUTTLE INTEGRATED VEHICLE (STS-1 CONFIGURATION)



(b) Integrated Vehicle.

Figure 3.5. - Concluded.

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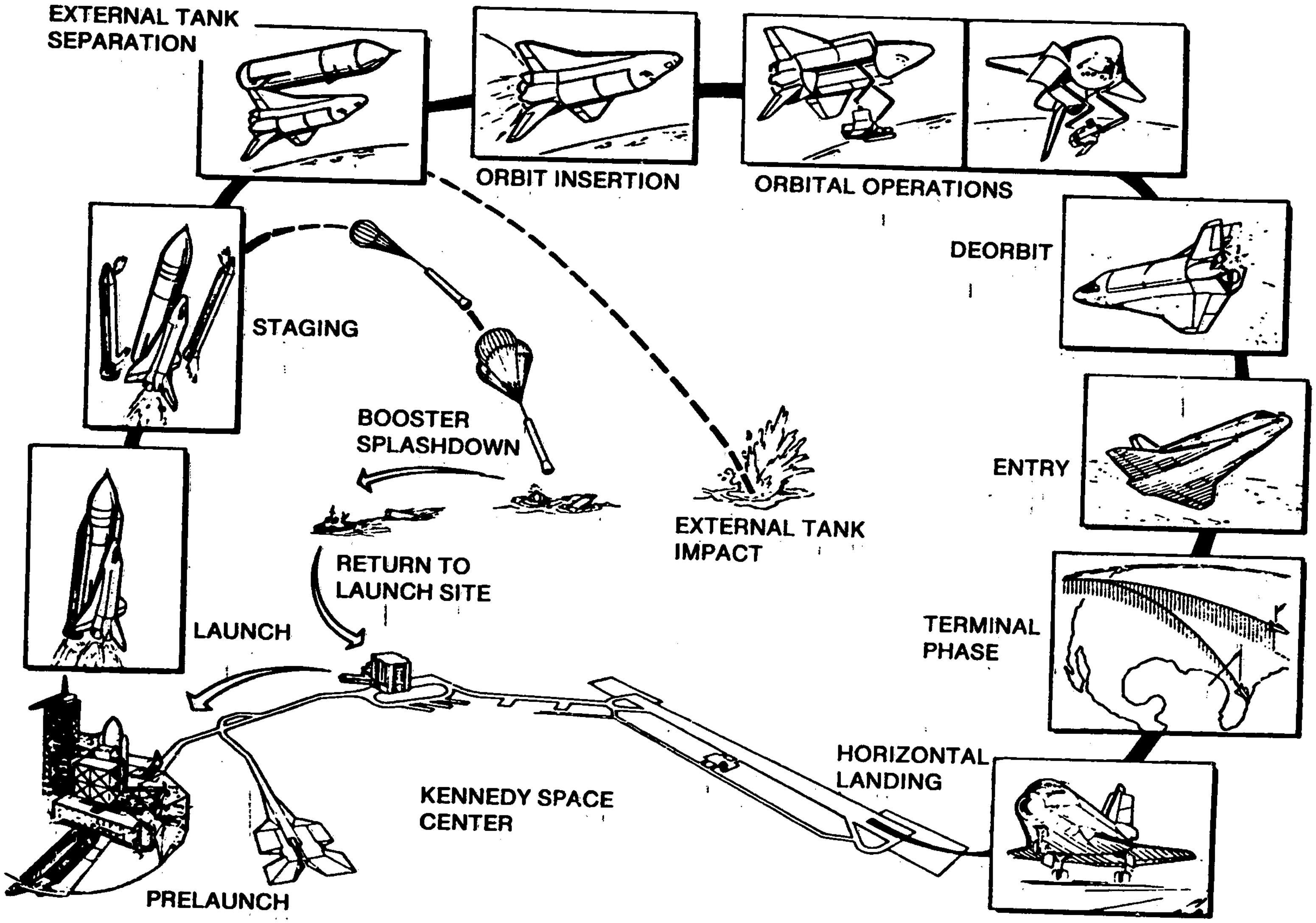
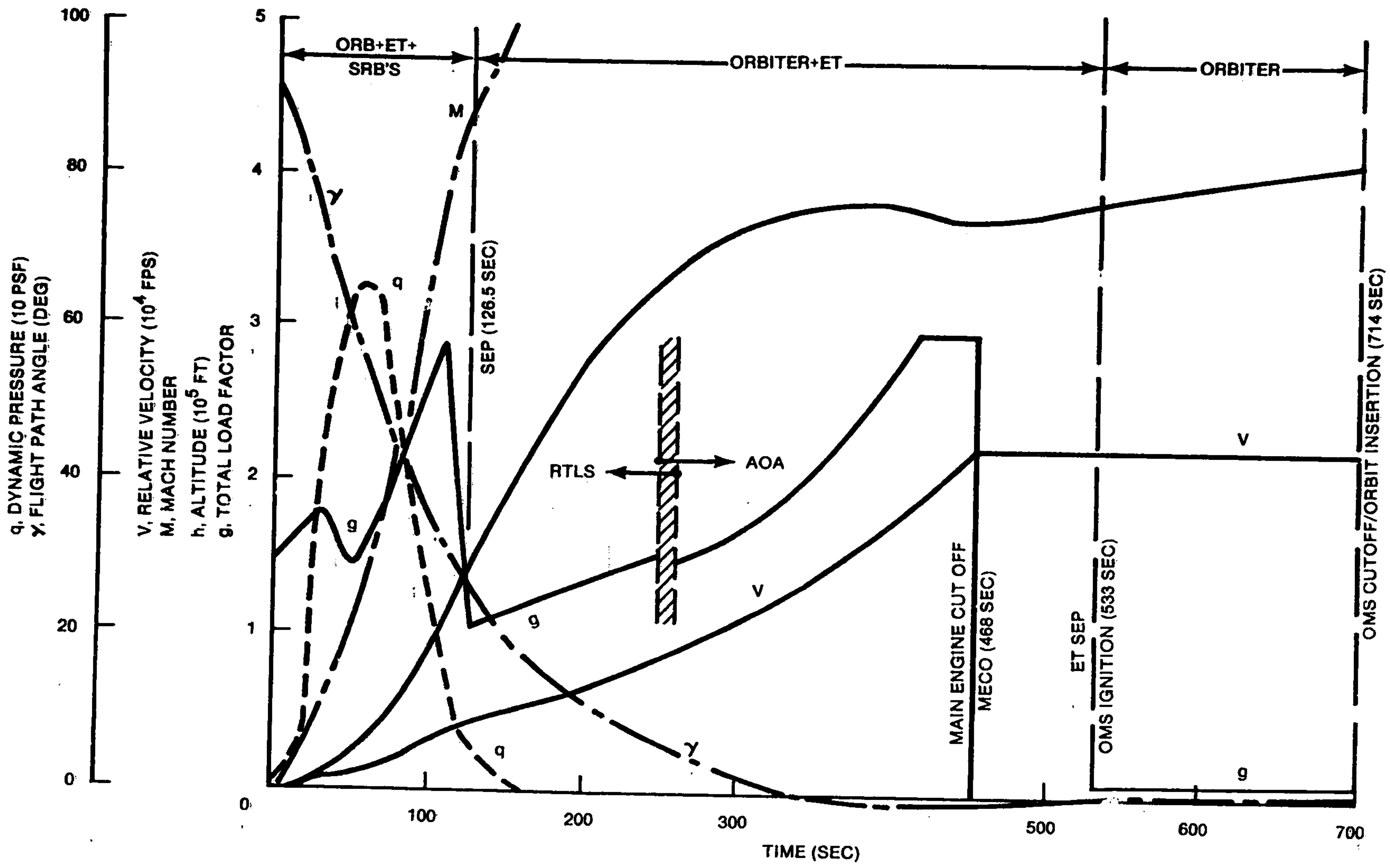
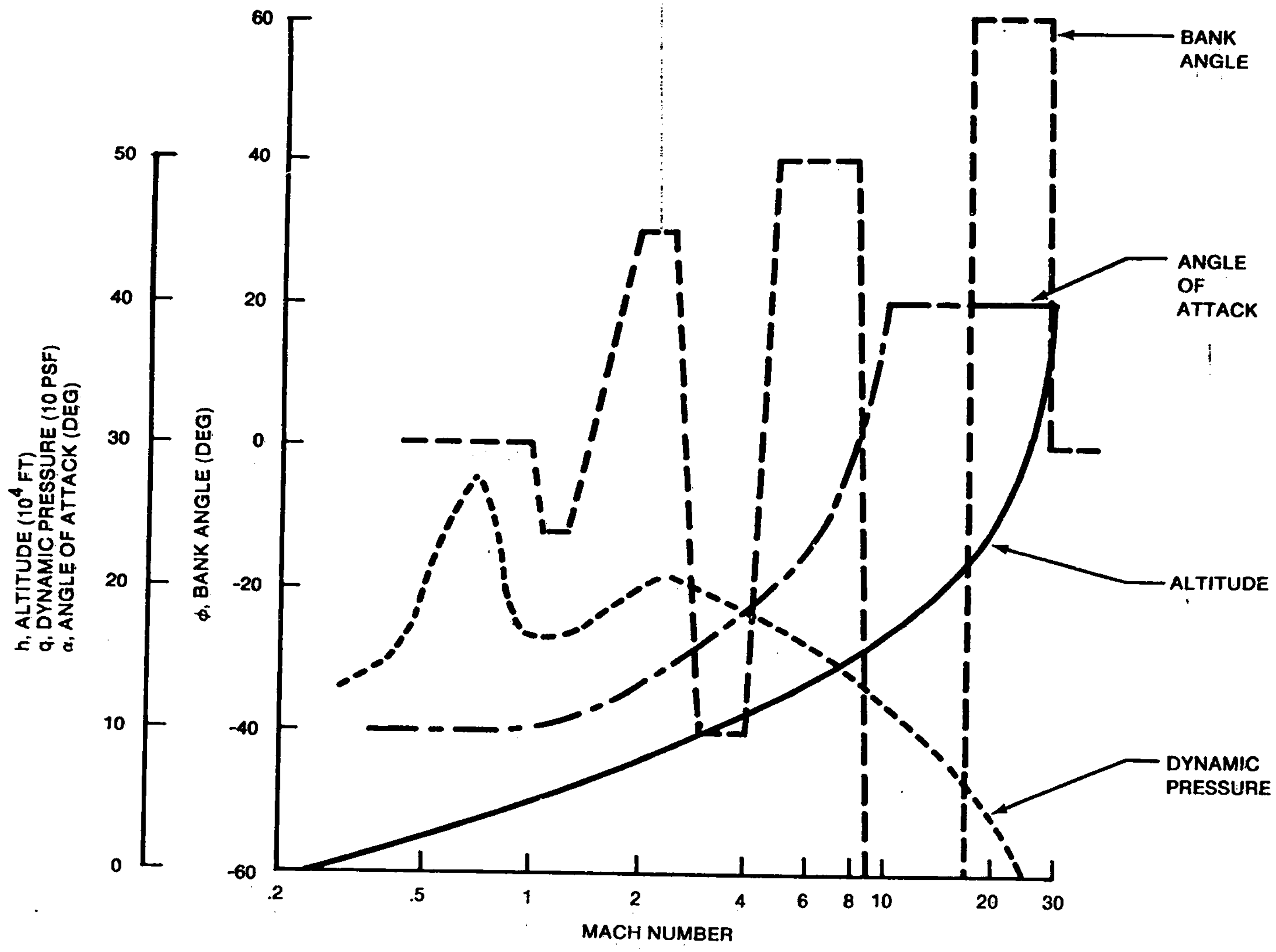


Figure 4.1. - Space Shuttle nominal mission phases.



(a) Ascent.

Figure 4.2. - Nominal trajectory characteristics.



(b) Entry.

Figure 4.2. - Concluded

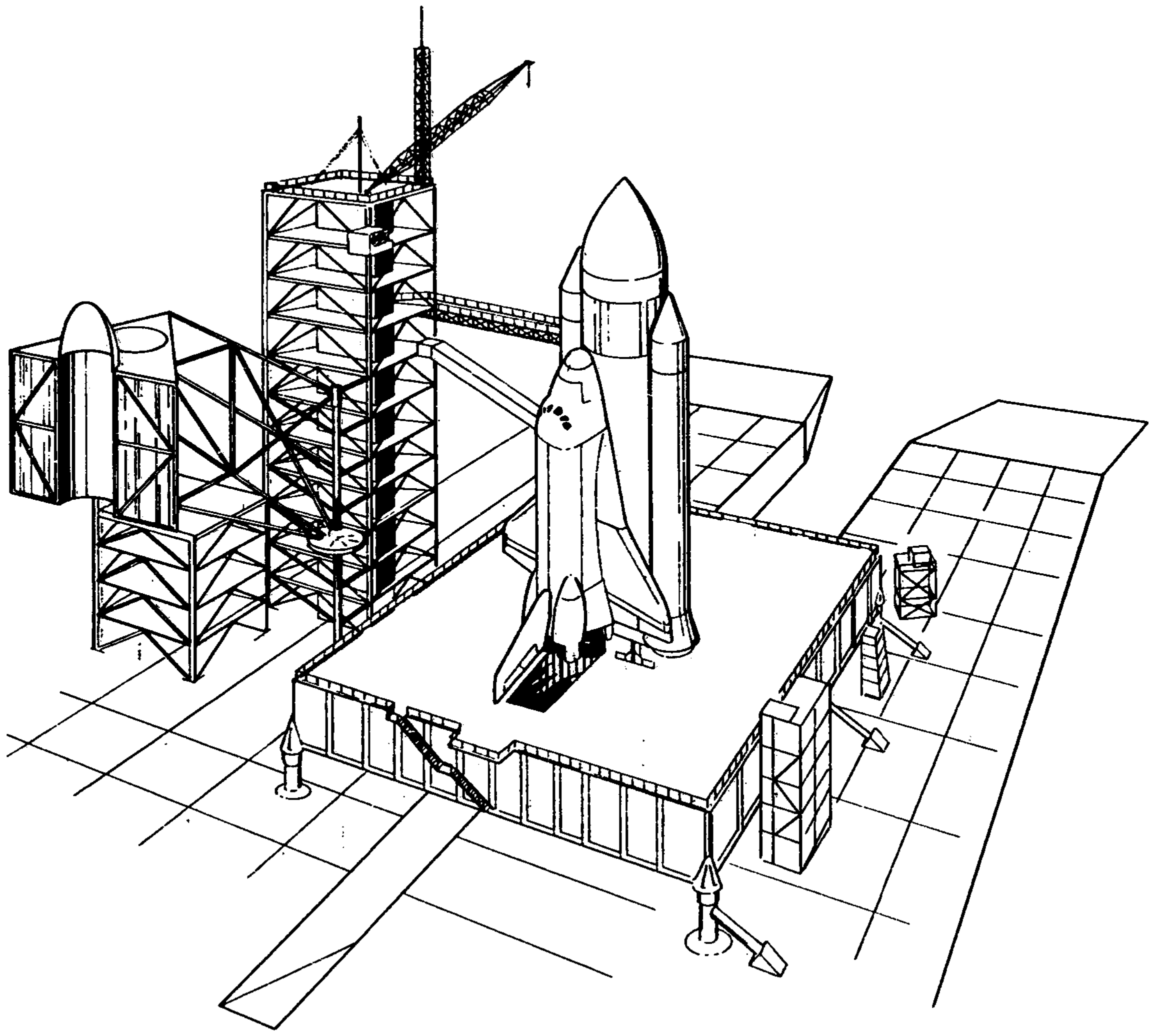


Figure 4.3. - Space Shuttle launch pad configuration.

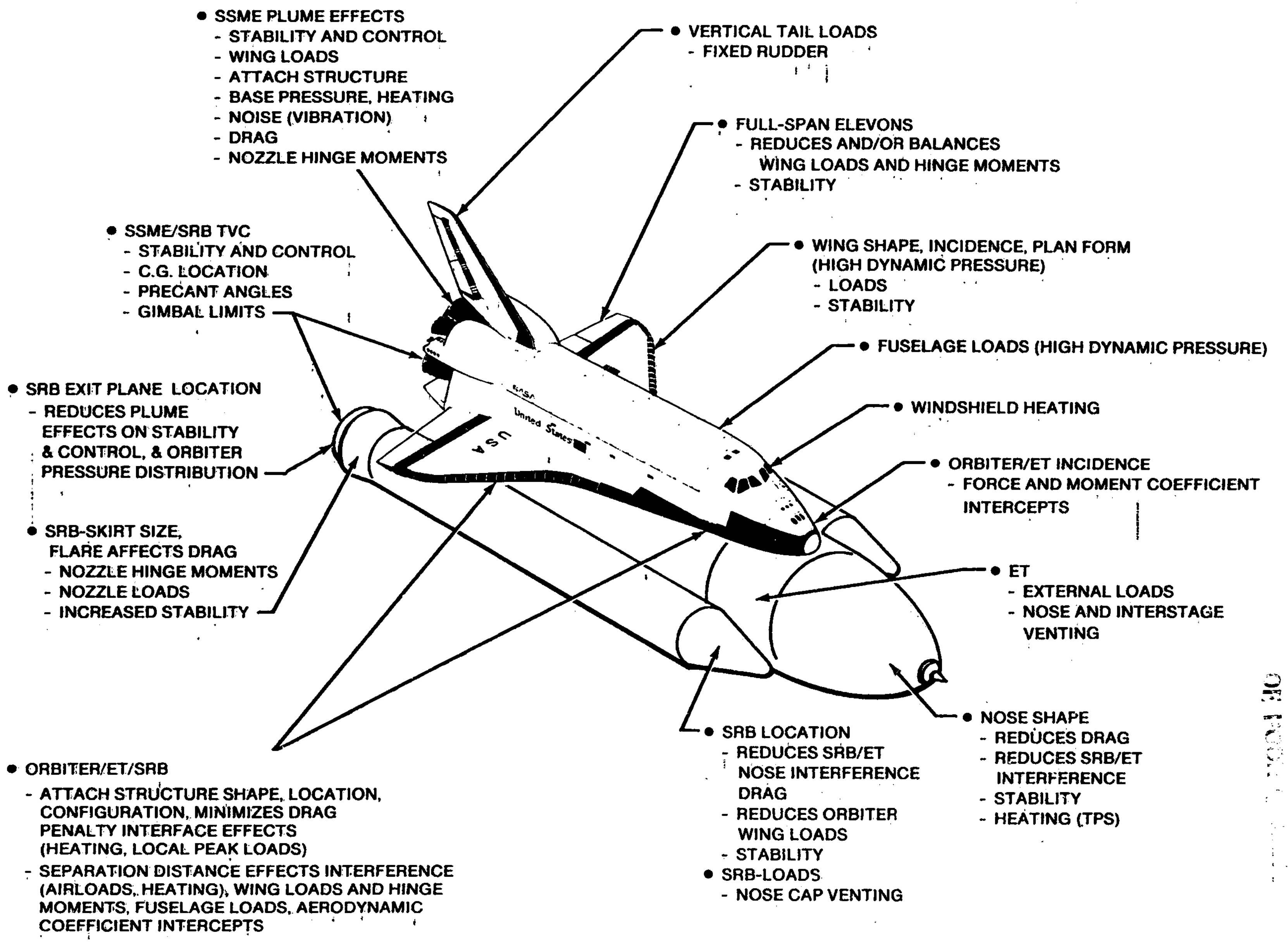


Figure 4.4. - Space Shuttle Integrated Vehicle aerodynamic considerations.

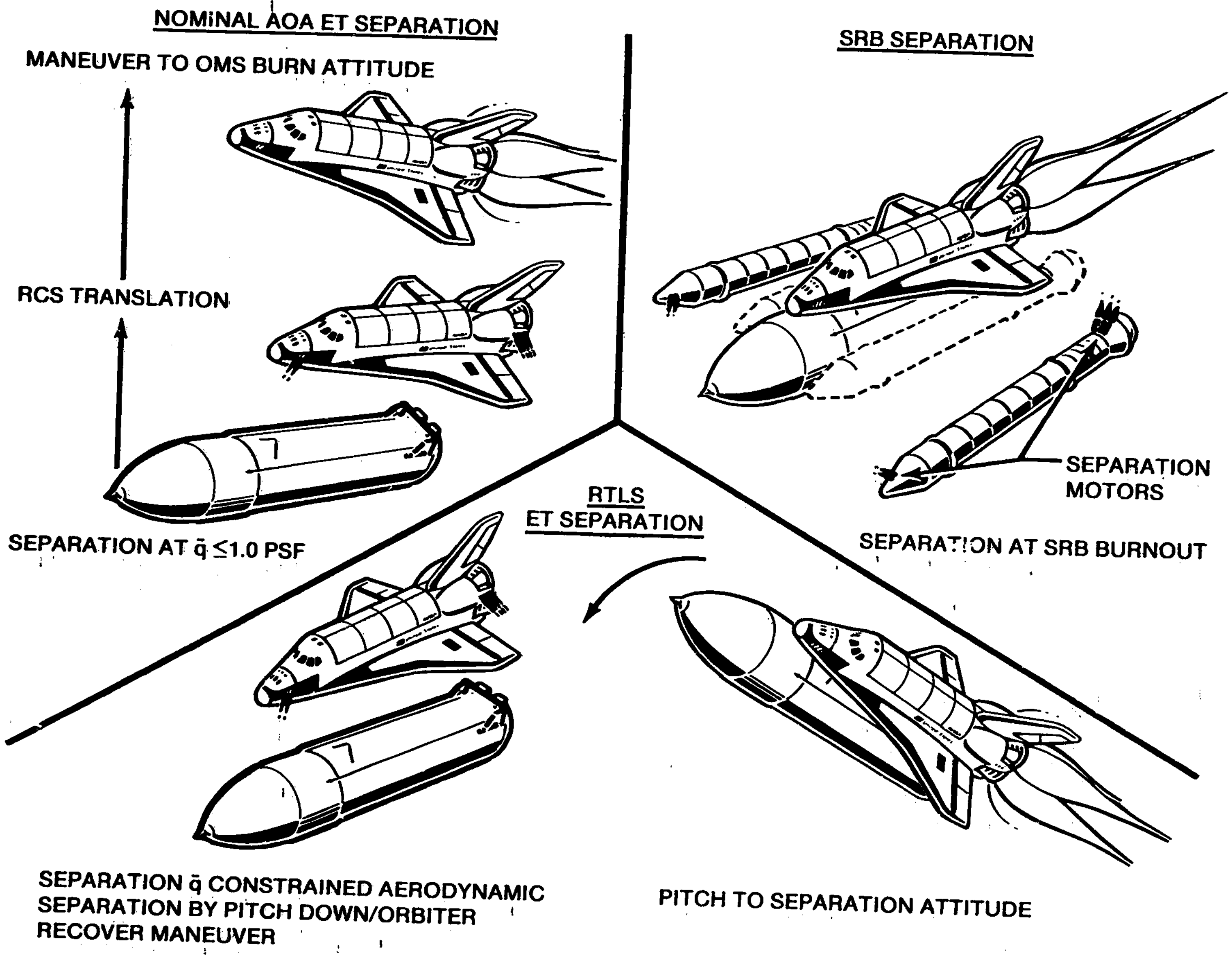


Figure 4.5. - Space Shuttle stage separation characteristics.

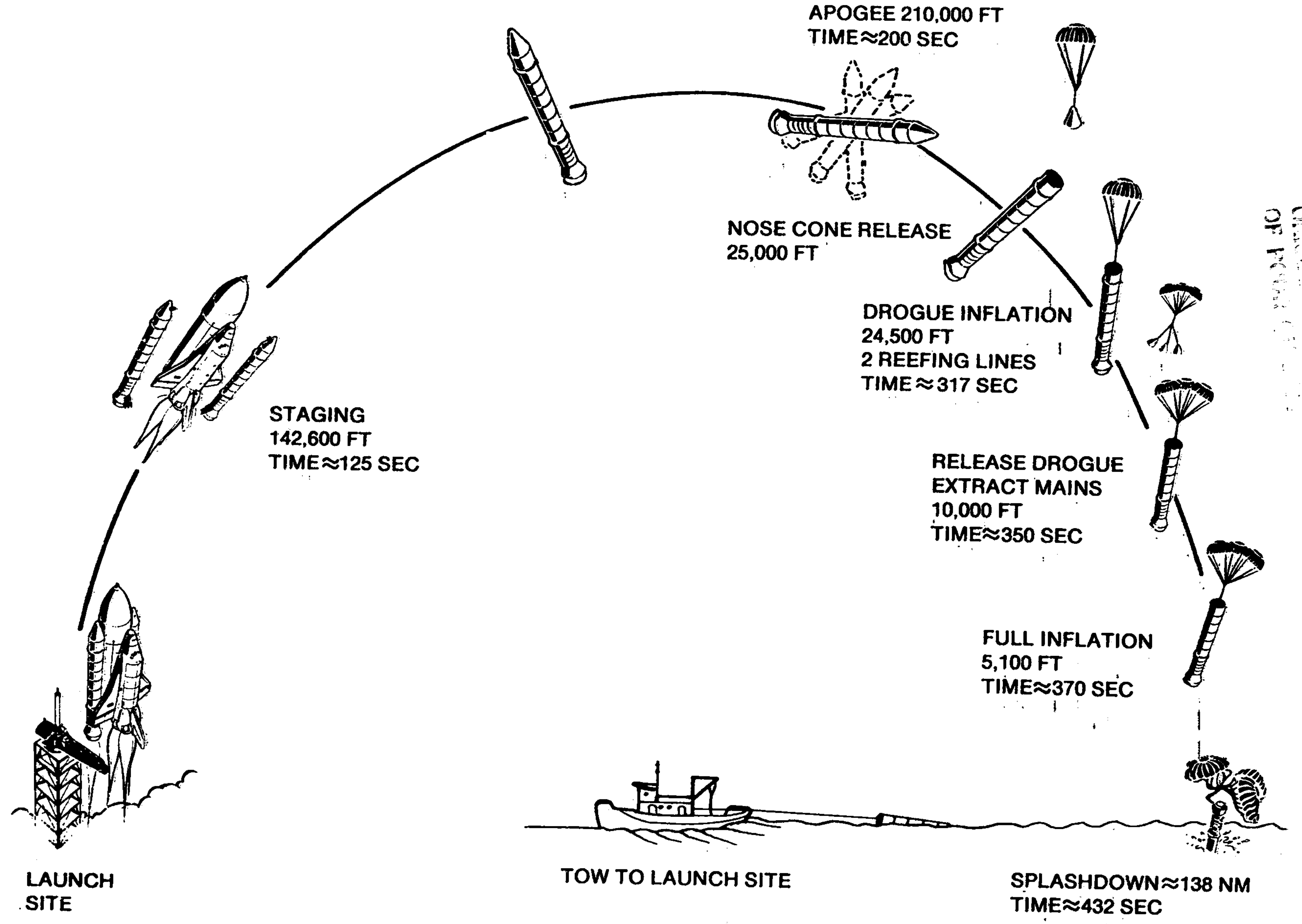


Figure 4.6. - Space Shuttle solid rocket booster recovery phases.

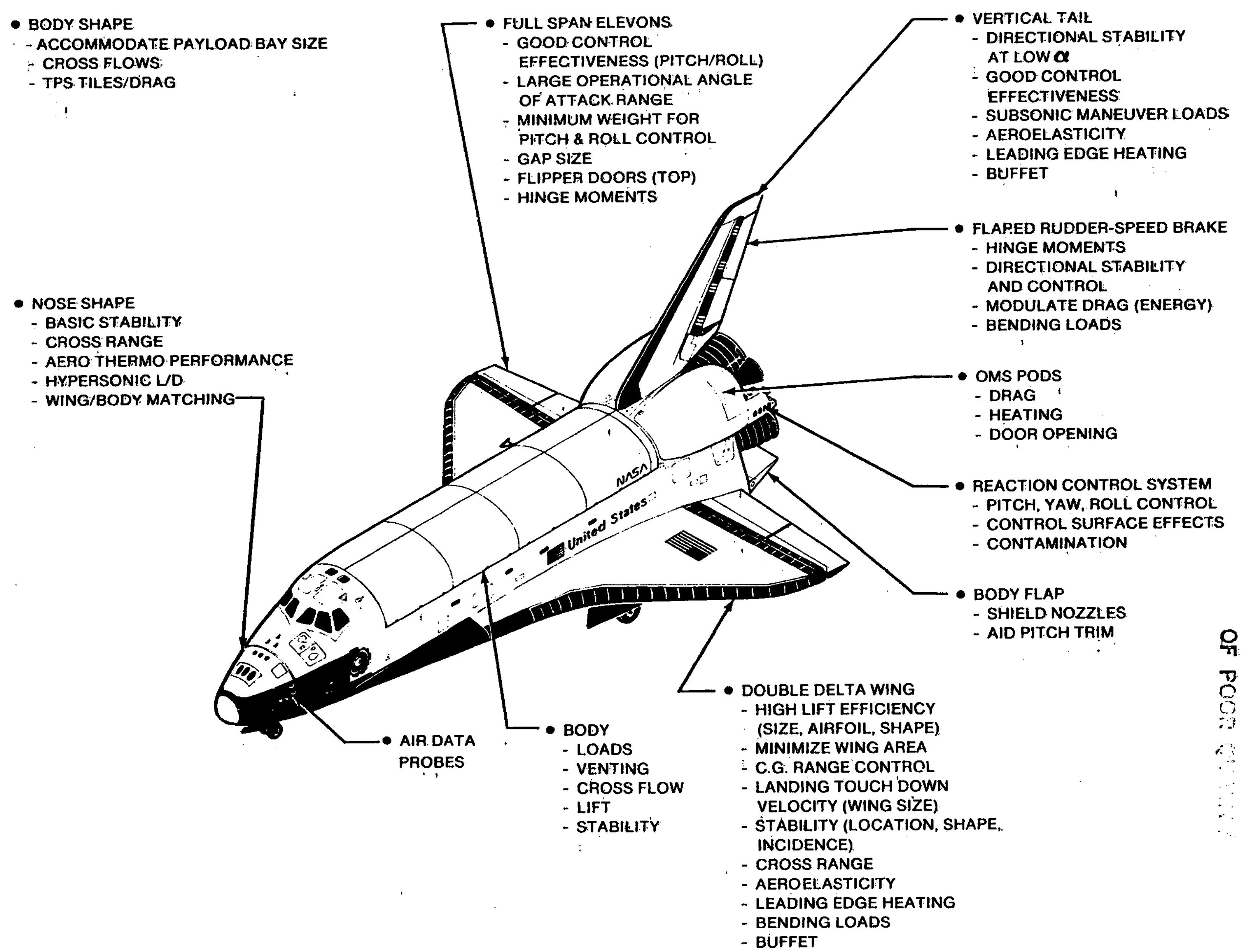


Figure 4.7. - Space Shuttle Orbiter Vehicle aerodynamic considerations.

	747 WING	747 VERT	747 HORIZ	ORBITER WING	ORBITER VERT
AREA	5500 FT^2 (1532.85 cm^2)	830 FT^2 (231.32 cm^2)	1470 FT^2 (409.69 cm^2)	2690 FT^2 (749.70 cm^2)	413.25 FT^2 (115.17 cm^2)
SPAN	2348 IN (5963.92 cm)	387 IN (982.98 cm)	864 IN (2194.56 cm)	936.68 IN (2379.17 cm)	315.72 IN (801.93 cm)
ASPECT RATIO	6.96	1.25	3.60	2.265	1.675
TAPER RATIO	0.356	0.340	0.250	0.200	0.404
SWEEP	37.5° (1/4C)	45.0°(1/4C)	37.5°(1/4C)	45°LE	45° LE
DIHEDRAL	7.0°	—	7.0°	3.5° TE	—
INCIDENCE	2.0°	—	+3° TO -12°	0.5°	—
MAC	327.78 IN. (832.56 cm)	334.16 IN. (848.77 cm)	271.60 IN. (689.86 cm)	474.72 IN. (1205.79 cm)	199.80 IN (489.51 cm)

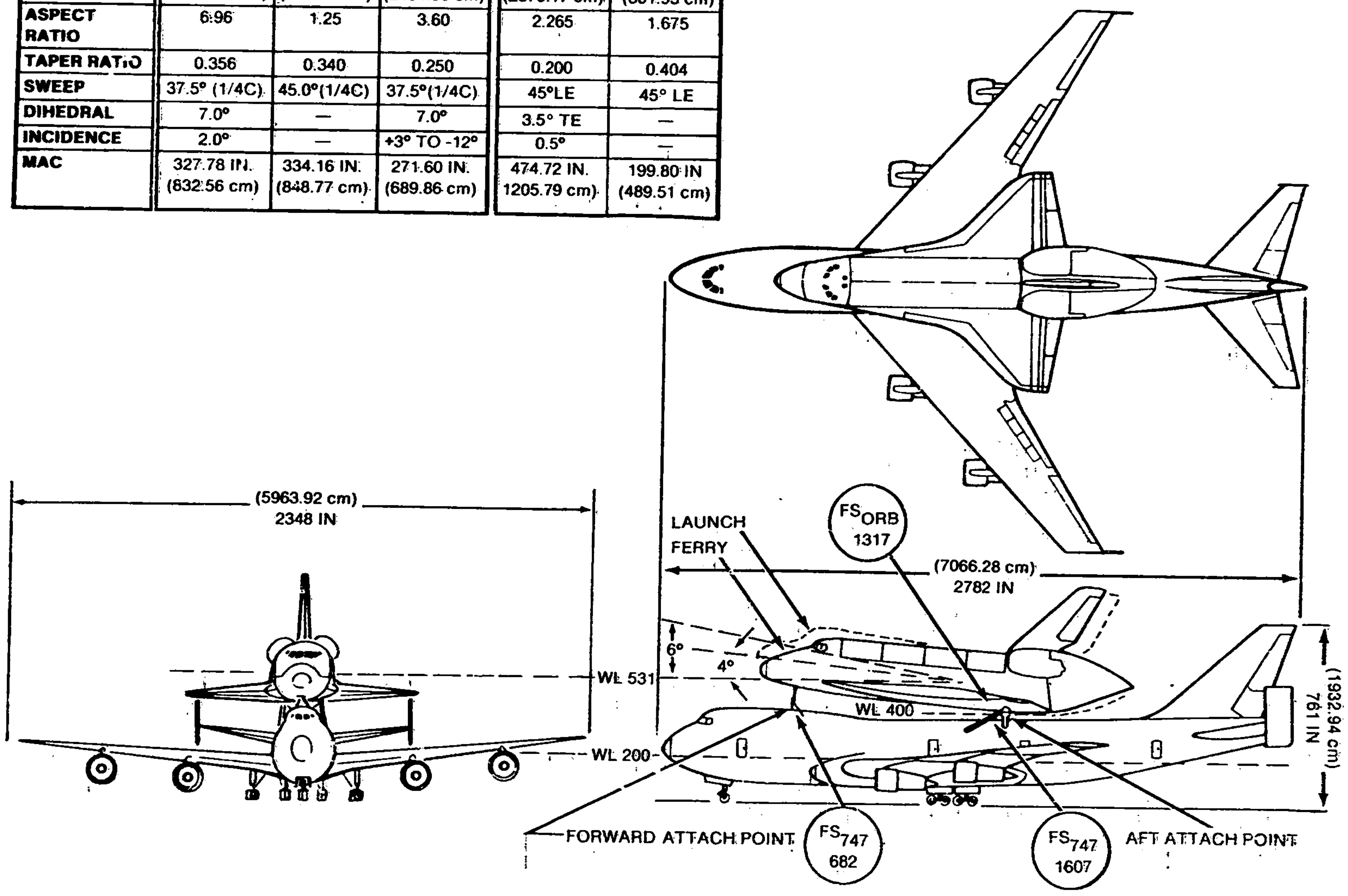


Figure 4.8. - Space Shuttle carrier aircraft mated configuration.

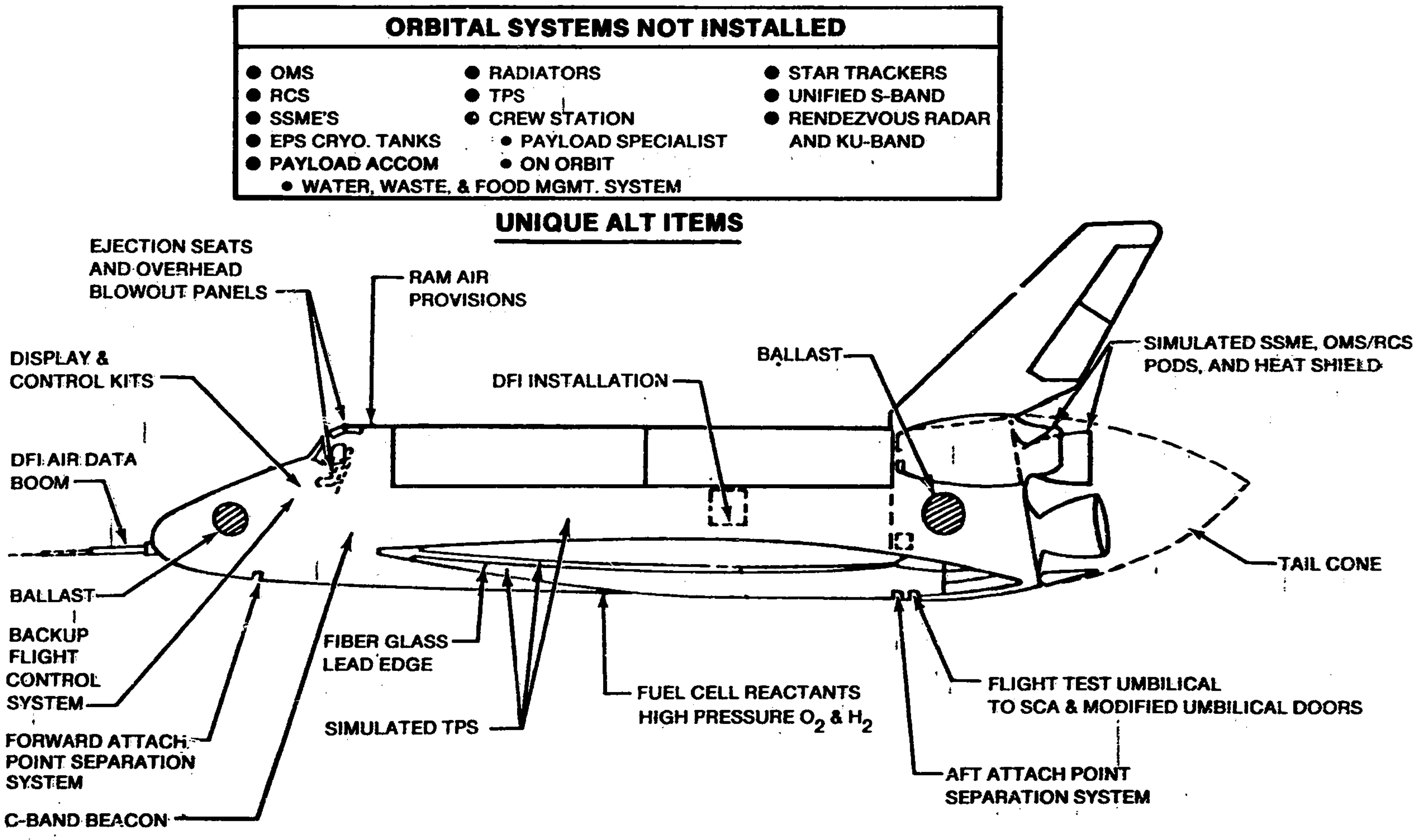


Figure 4.9. - Space Shuttle Orbiter Vehicle-101 configuration for ALT.

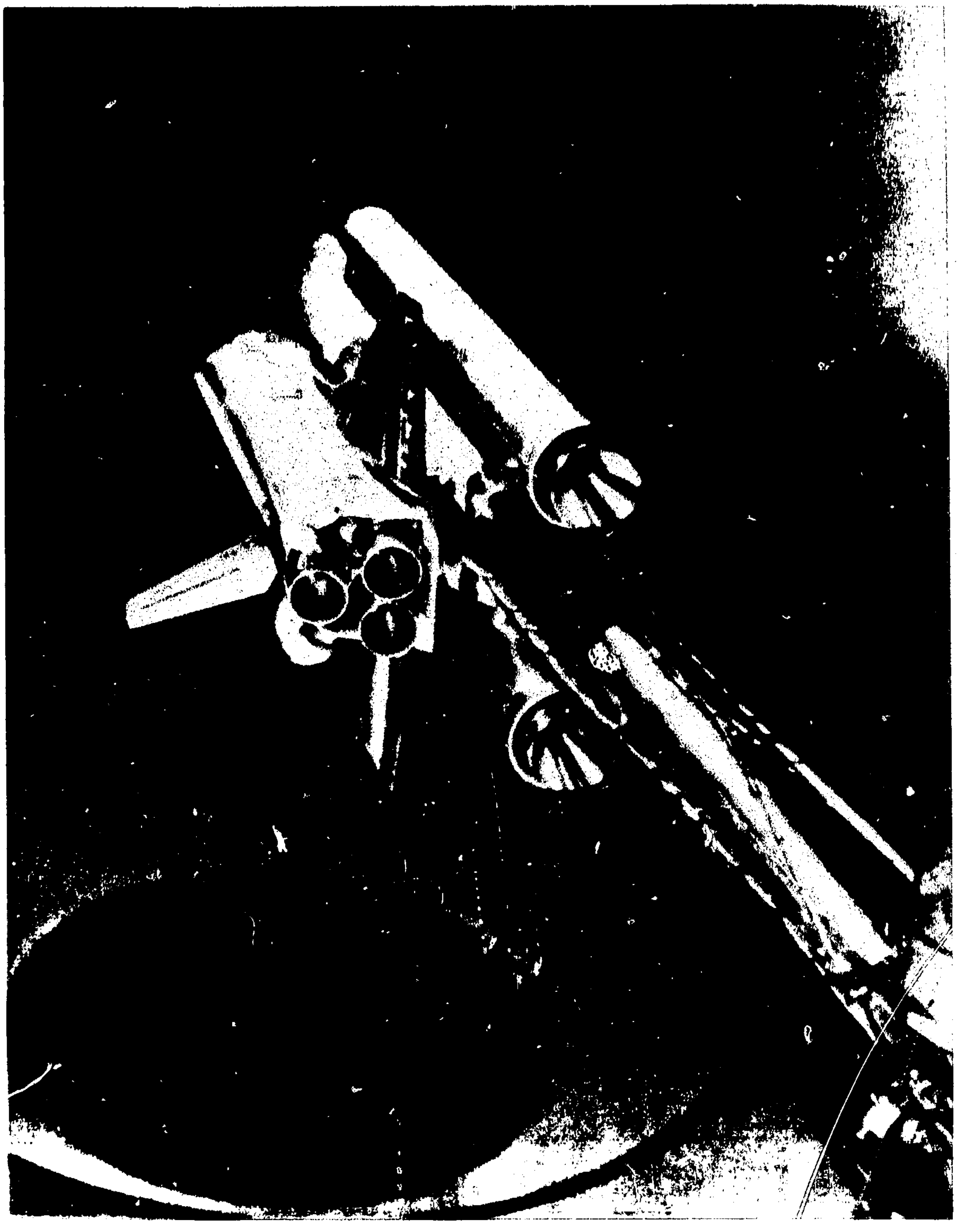


Figure 5.1. - Integrated Vehicle no. 14 sting-supported from ET base; ARC 8- by 7-ft Supersonic Wind Tunnel.

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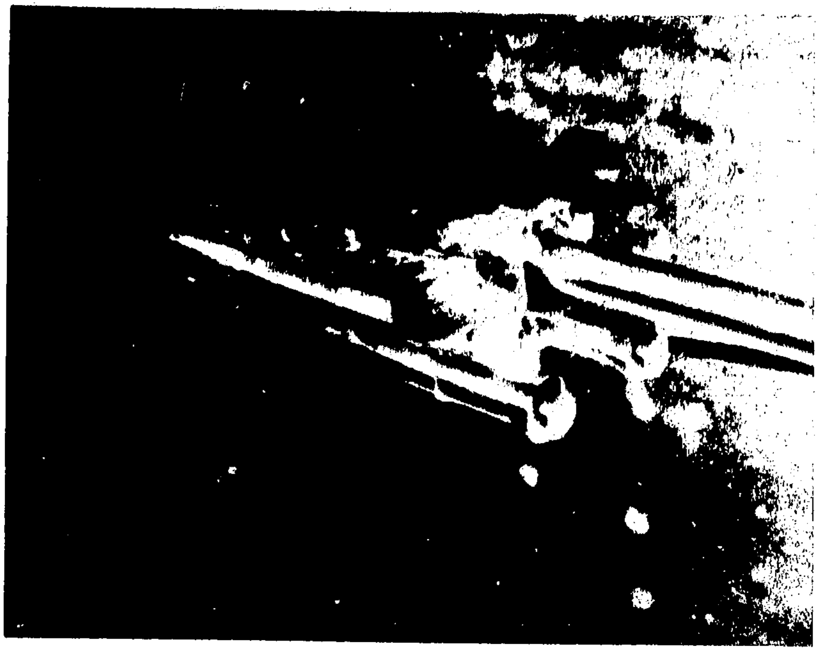
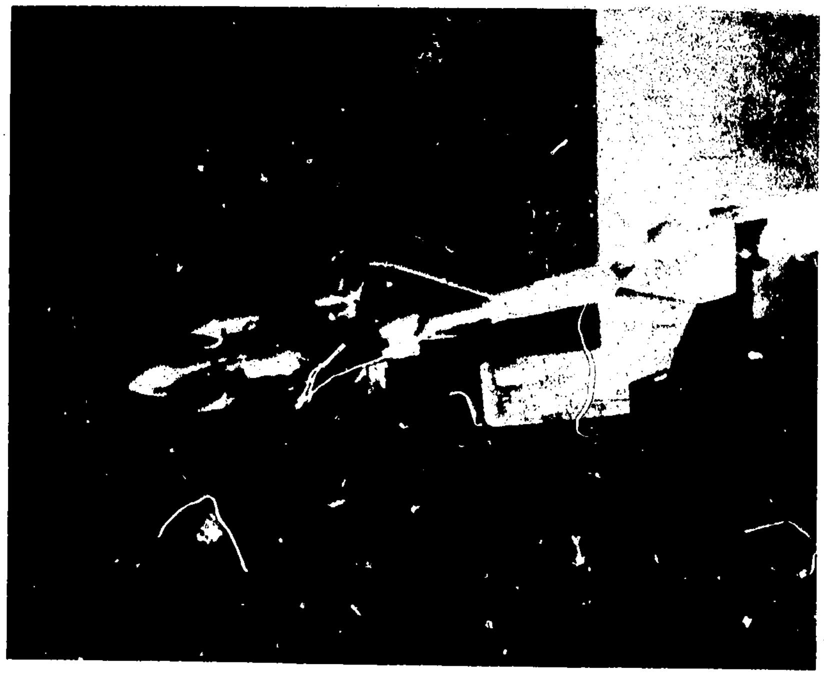
Figure 5.2. - Integrated Vehicle model no. 67 sting-supported from Orbiter base; LaRC 8-ft Transonic Pressure Tunnel.

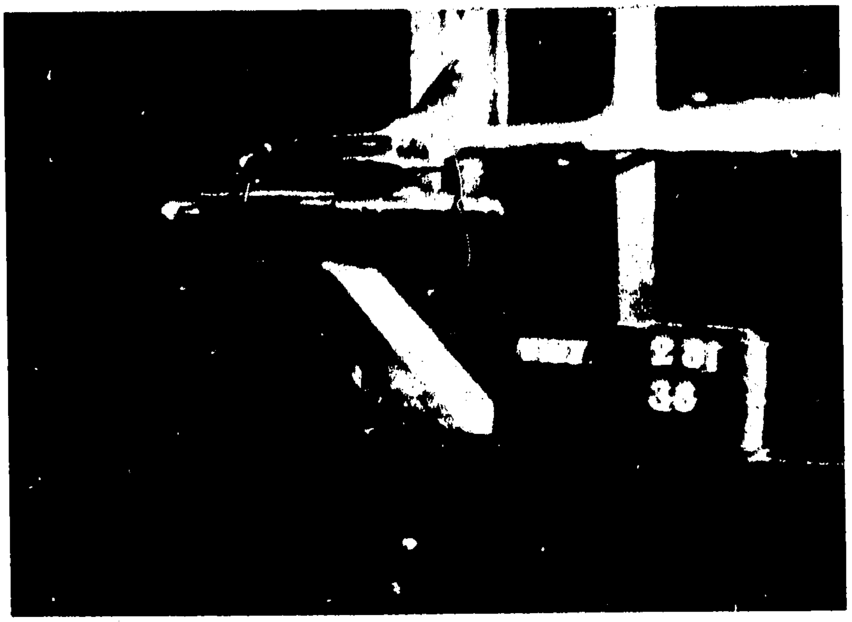


(a) Sting support from Orbiter base.

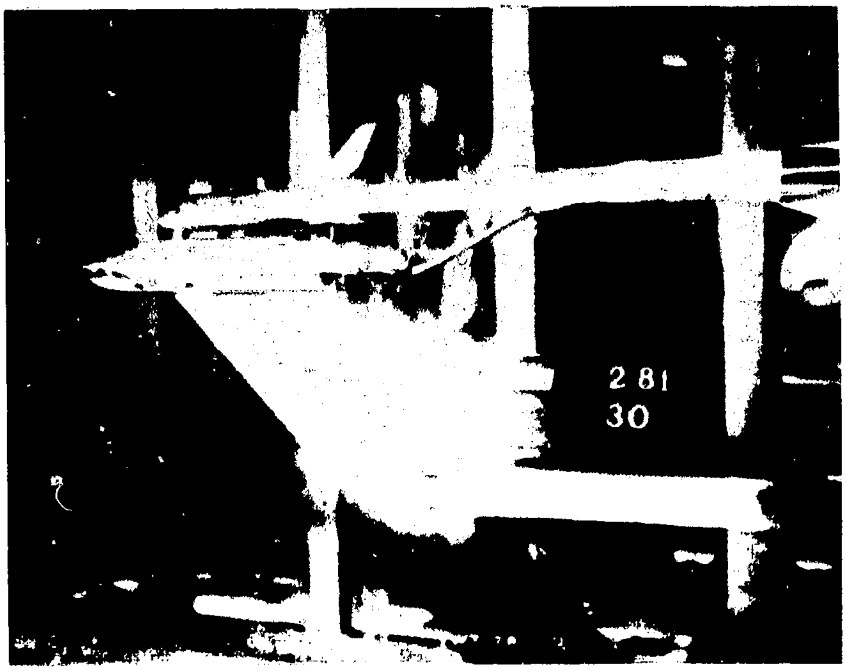


(b) Blade support from SRB side with dummy sting.

Figure 5.3. - Integrated Vehicle model support interference tests; model 13 in the RI 7-ft Trisonic Wind Tunnel.

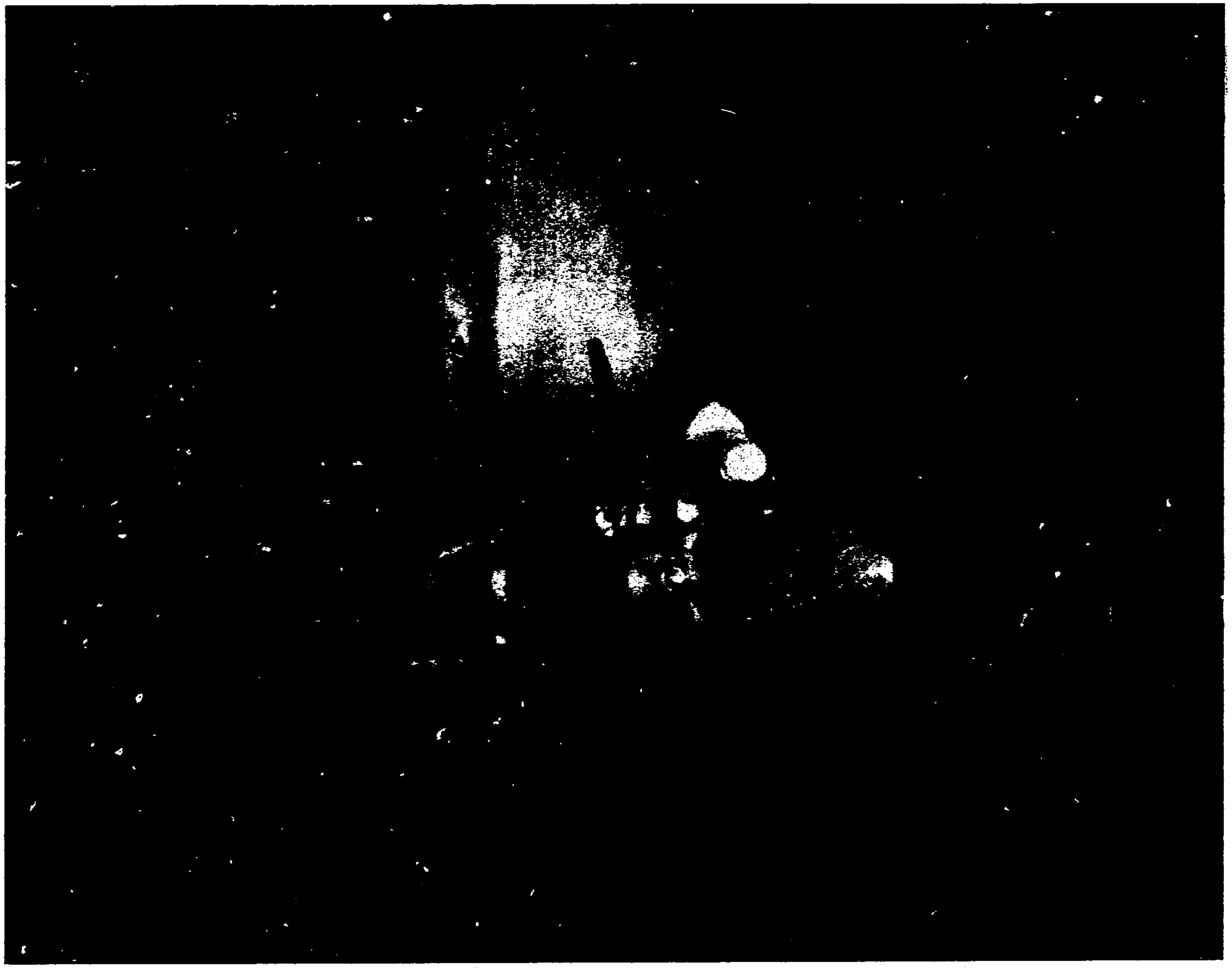


(c) Blade support from ET bottom (forward position) with dummy sting.



(d) Blade support from ET bottom (aft position) with dummy sting.

Figure 5.3. - Concluded.



(a) Model no. 14 in the ARC 8- by 7-ft Supersonic Wind Tunnel.

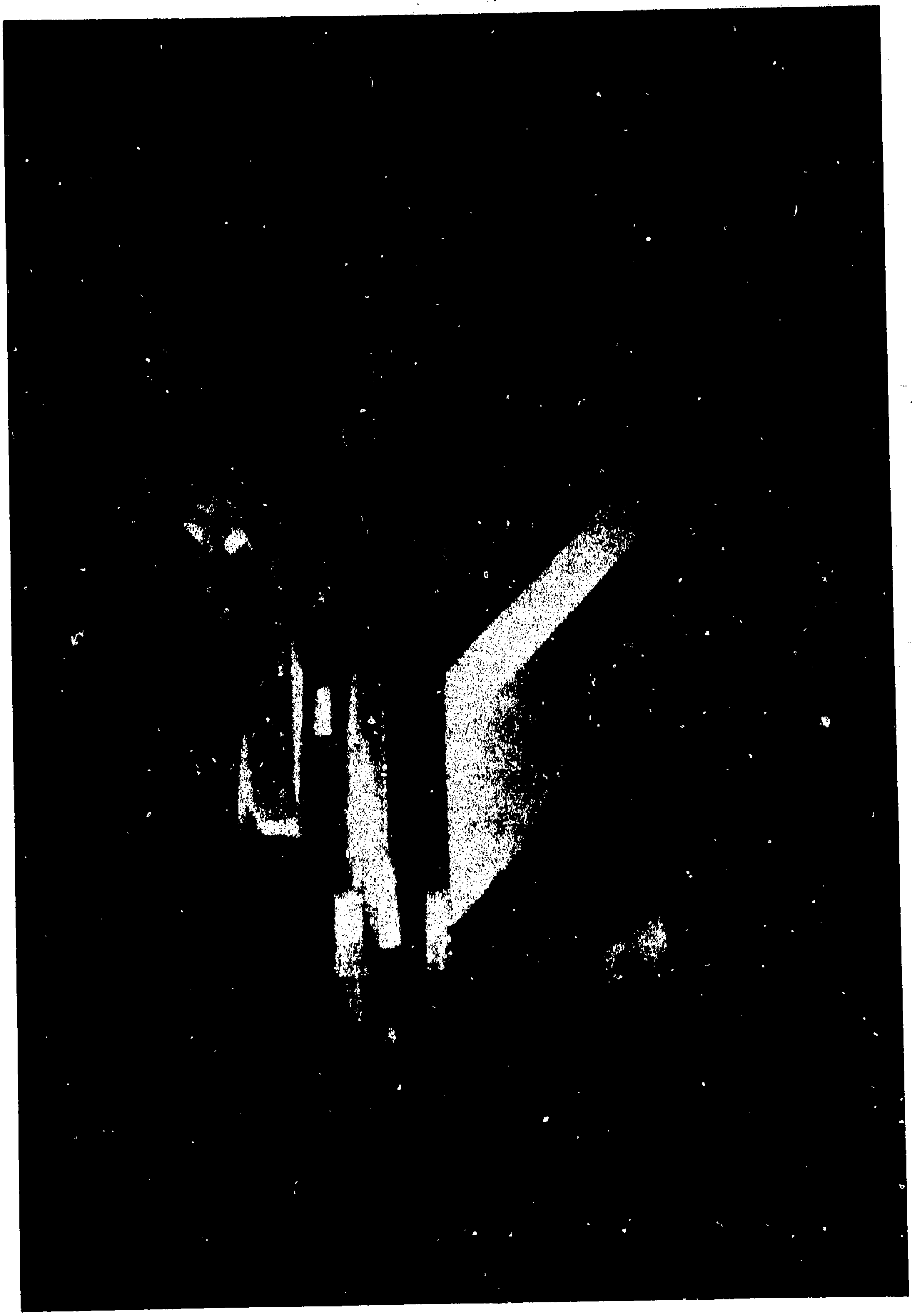
Figure 5.4. - Integrated Vehicle plume effects tests; sting-supported.

(b) Model no. 14 in the ARC 9- by 7-ft Supersonic Wind Tunnel.

Figure 5.4. - Concluded.

(a) Front view.

Figure 5.5. - Integrated Vehicle plume effects tests; blade-supported model no. 88 in the ARC 11-ft Transonic Wind Tunnel.



(b) Side view with active plumes.

Figure 5.5. - Continued.

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(c) Base view.

Figure 5.5. - Concluded.

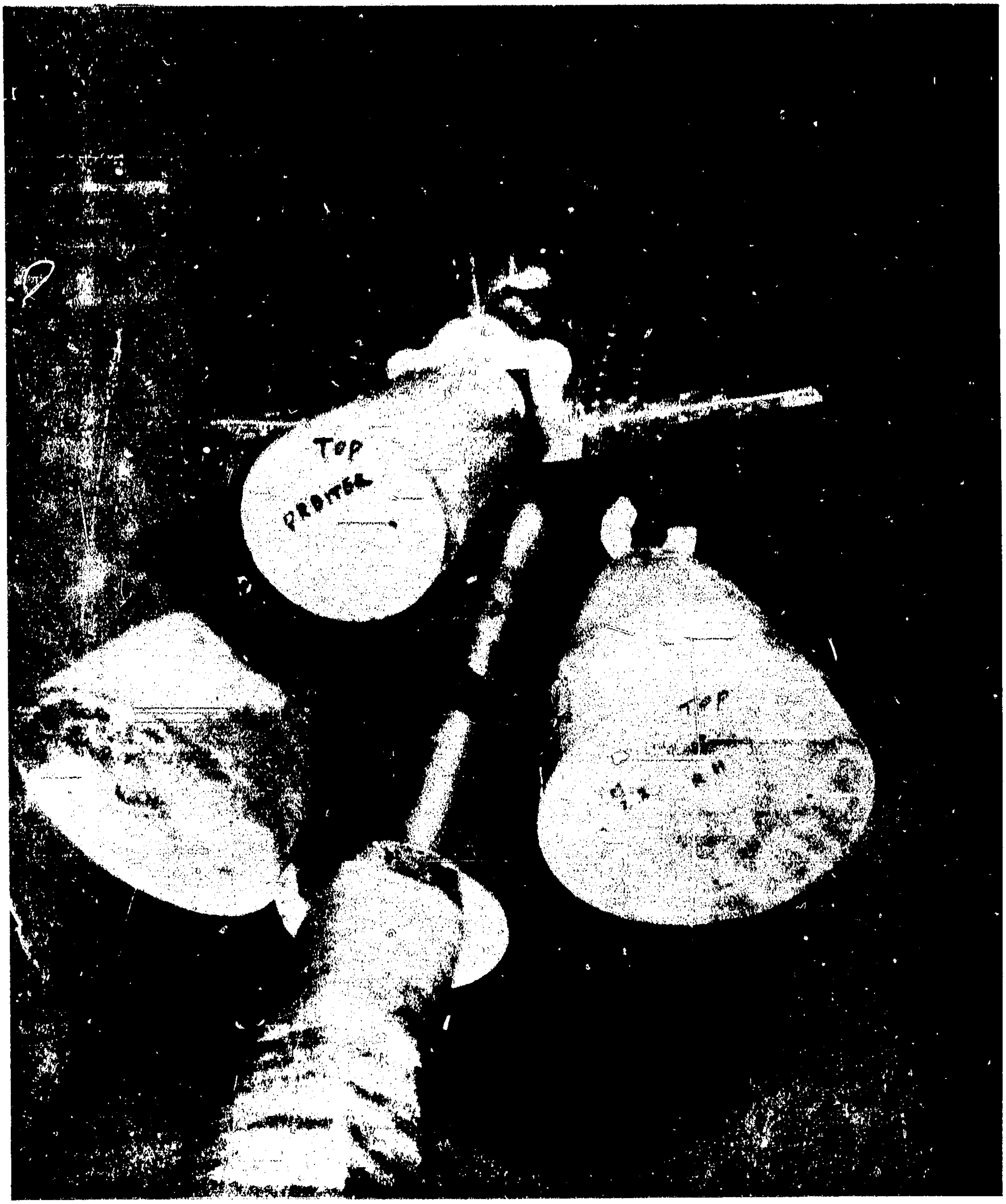

Figure 5.6. - Integrated Vehicle plume effects test; solid plumes, model no. 14 in the ARC 9- by 7-ft Supersonic Wind Tunnel.

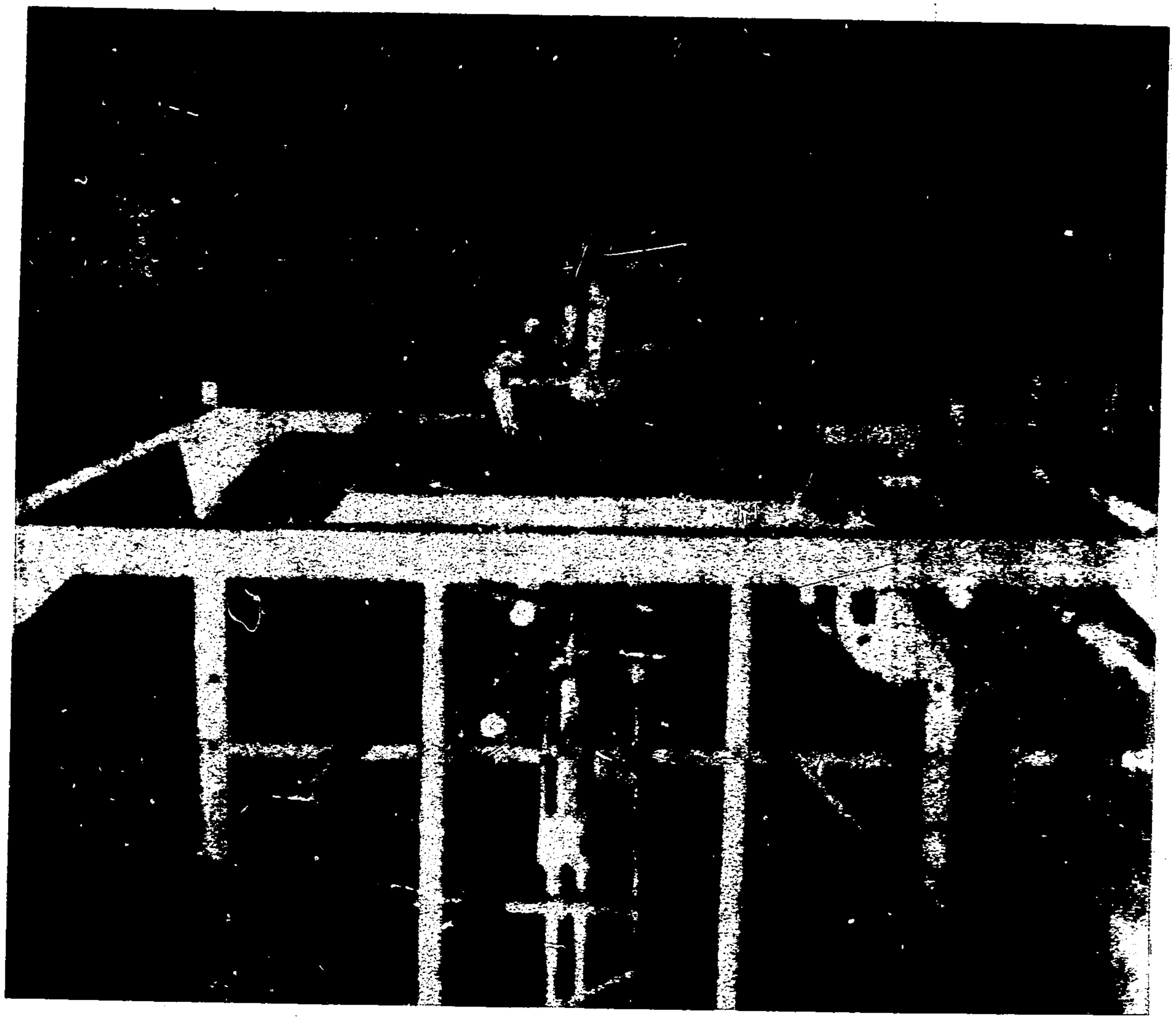

Figure 5.7. - Integrated Vehicle plume effects at high altitudes; model no. 25 in the JSC Vacuum Chamber Facility.

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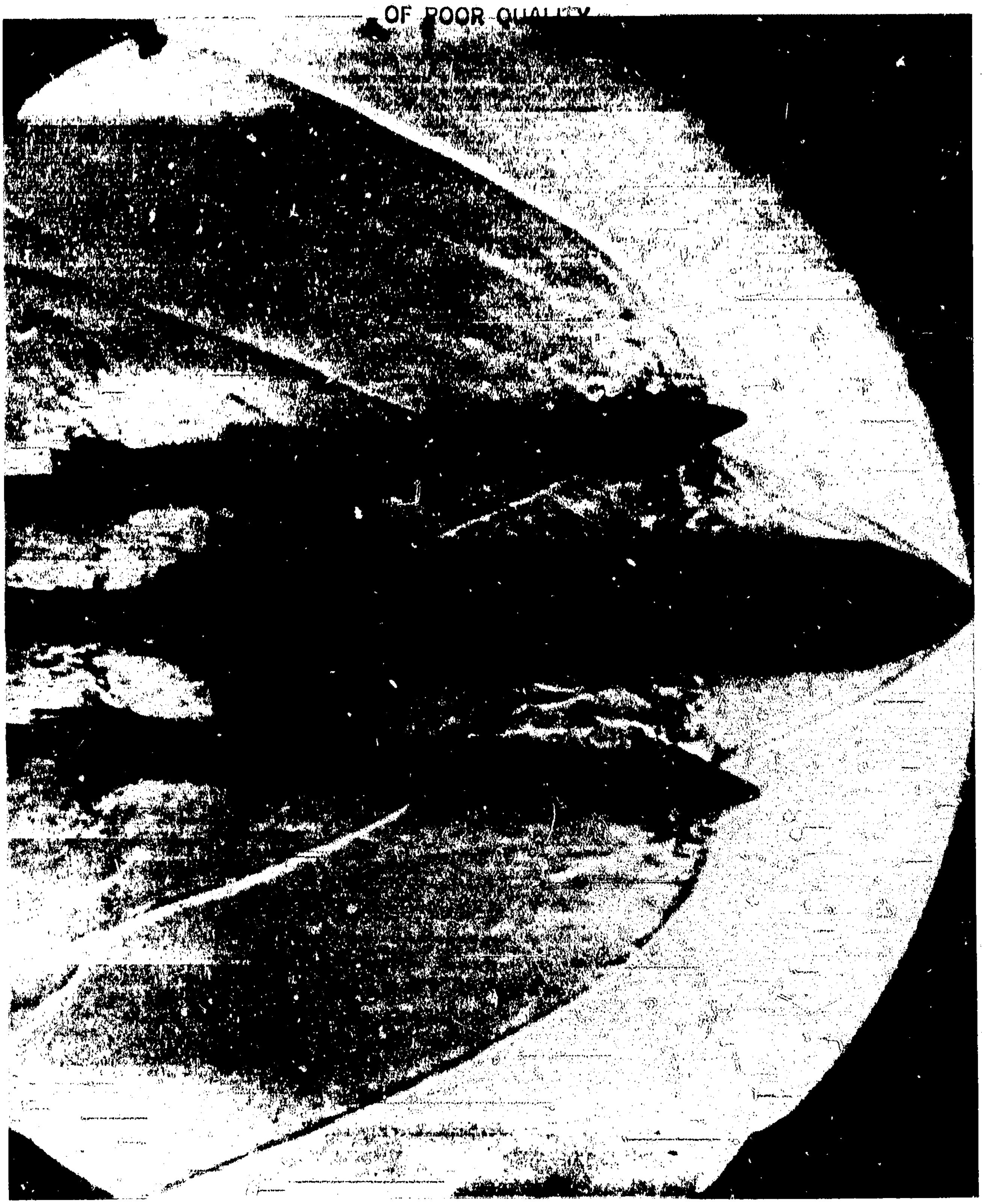


(a) Plan view.

Figure 5.8. - First stage separation test with SRB separation-motors simulated; model no. 32 in the AEDC-A Tunnel.

(b) Right side view.

Figure 5.8. - Concluded.

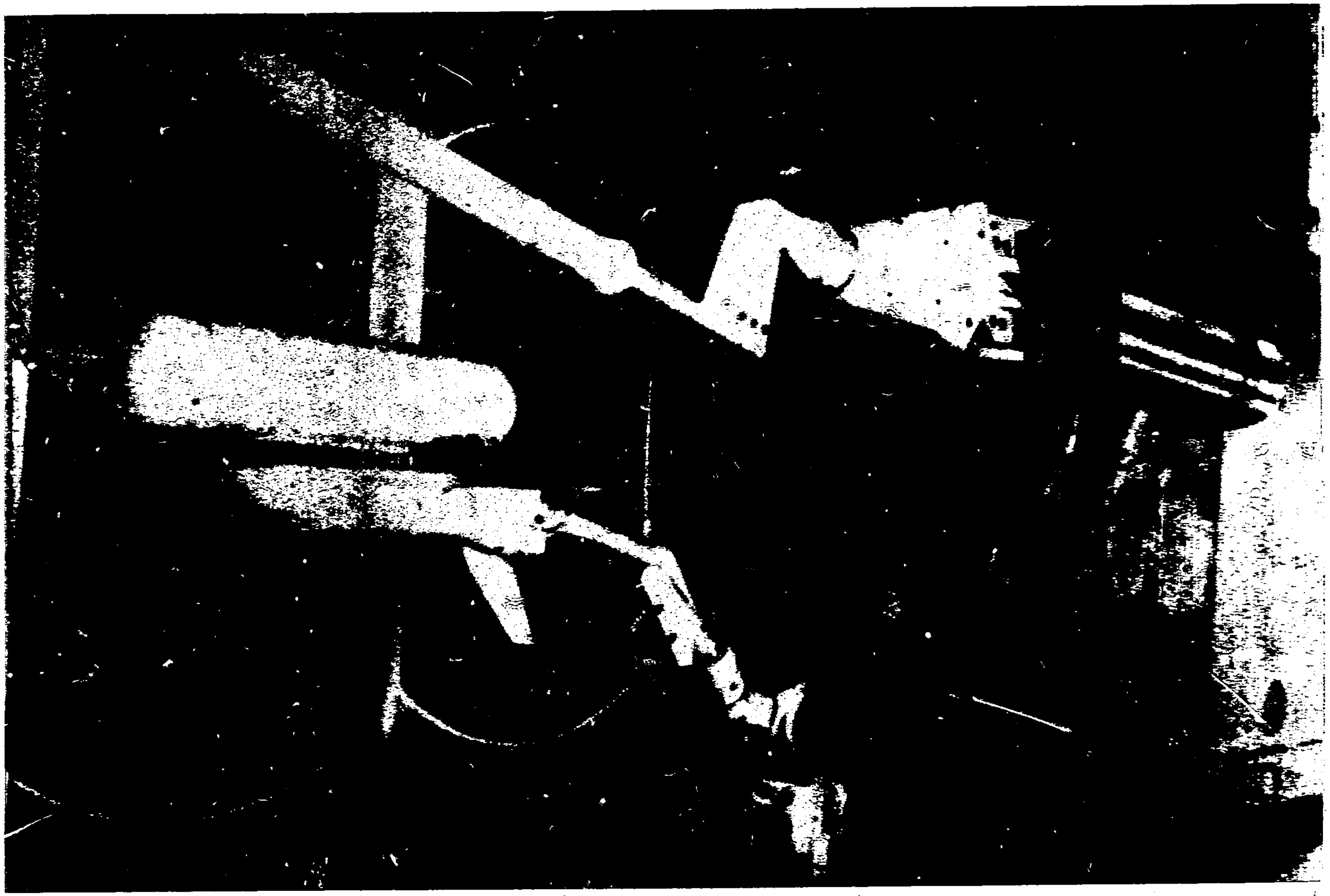


Figure 5.9. - First stage separation test (motor off); model no. 52 using the captive trajectory system in the AEDC-A Tunnel.

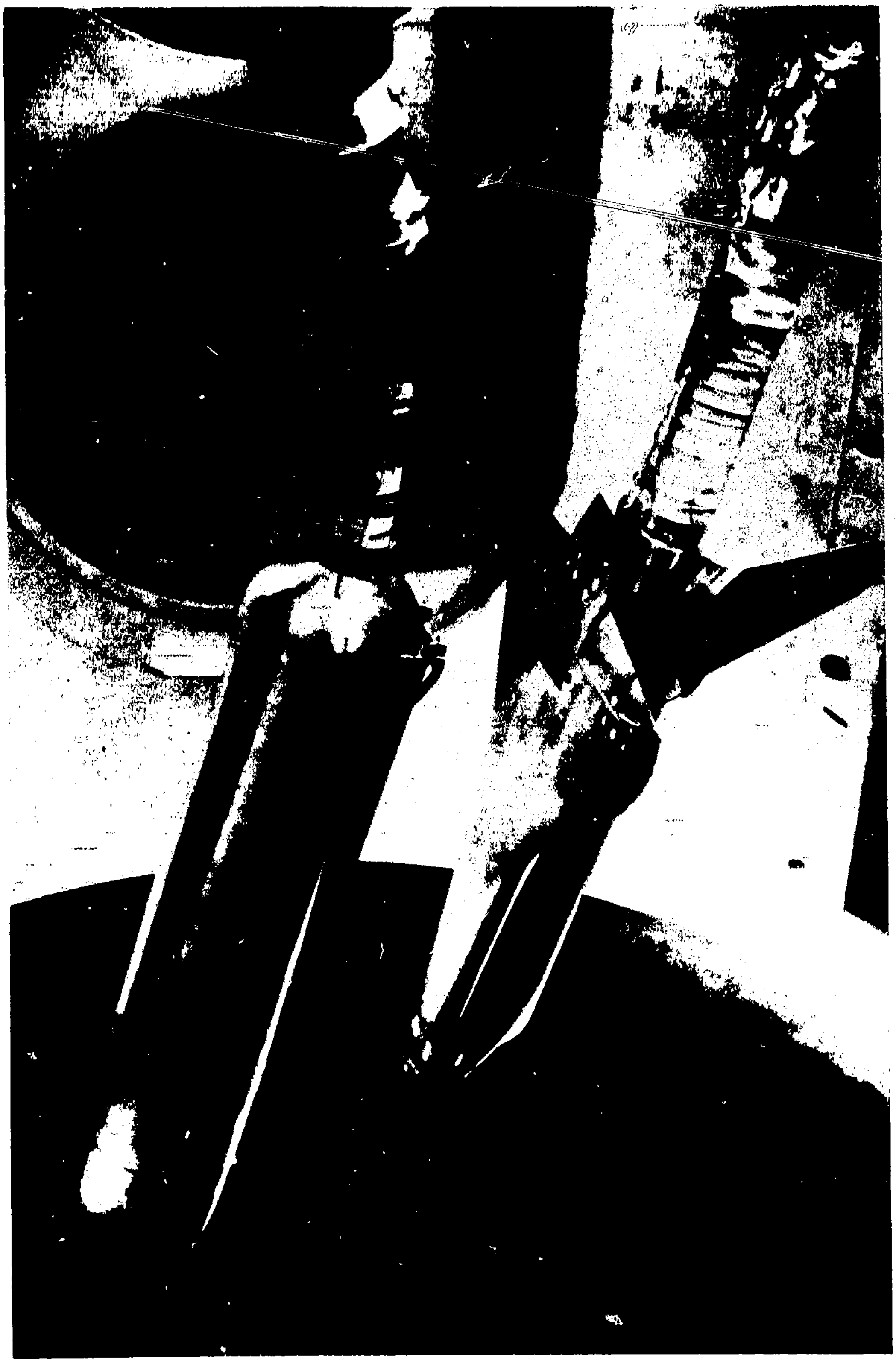


Figure 5.10. - Second stage separation test (RTLS conditions); model no. 70 in the AEDE-B Tunnel.

C-2

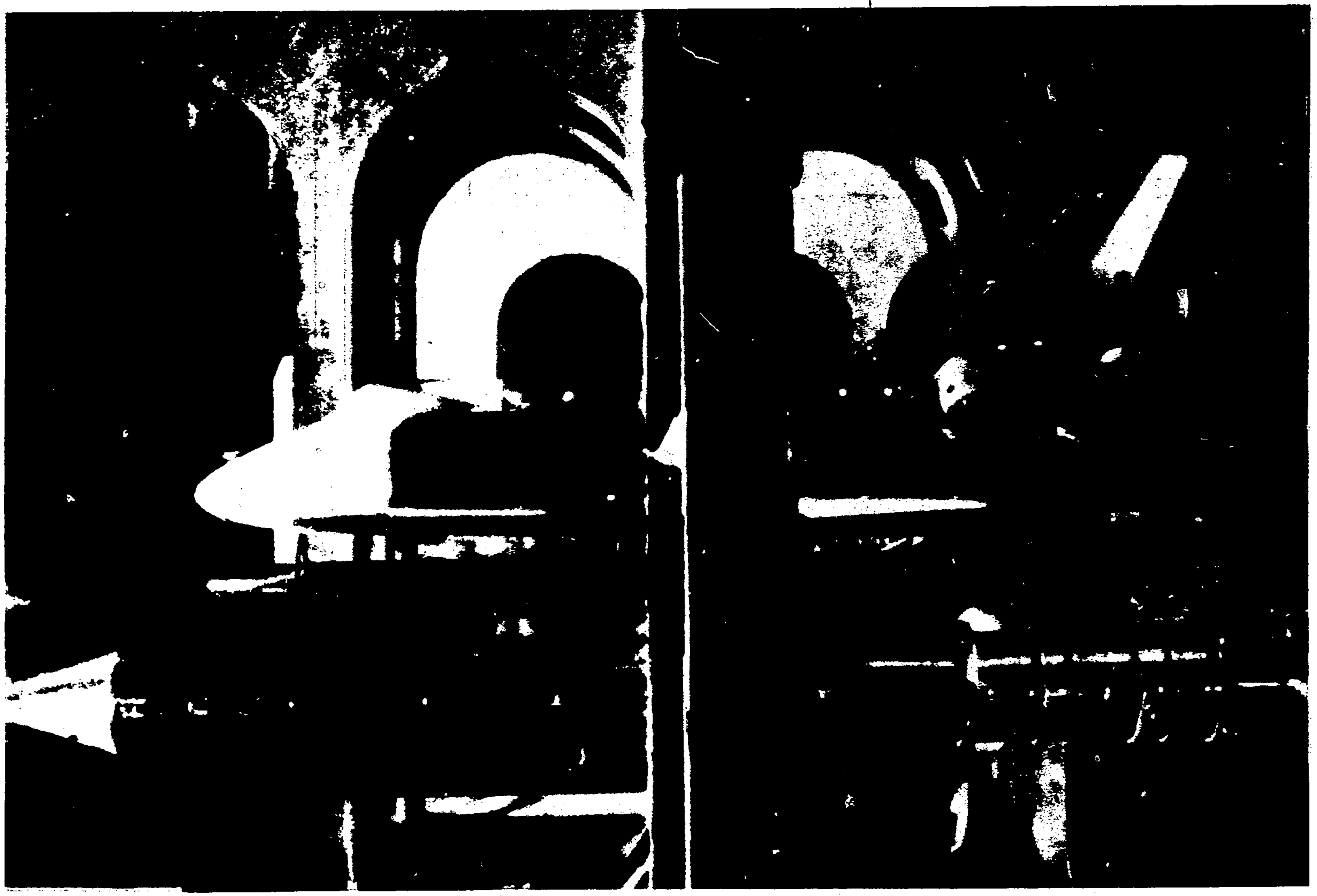


Figure 5.11. - Heating test using thermocouples; model no. 60 in the AEDC-A Tunnel.

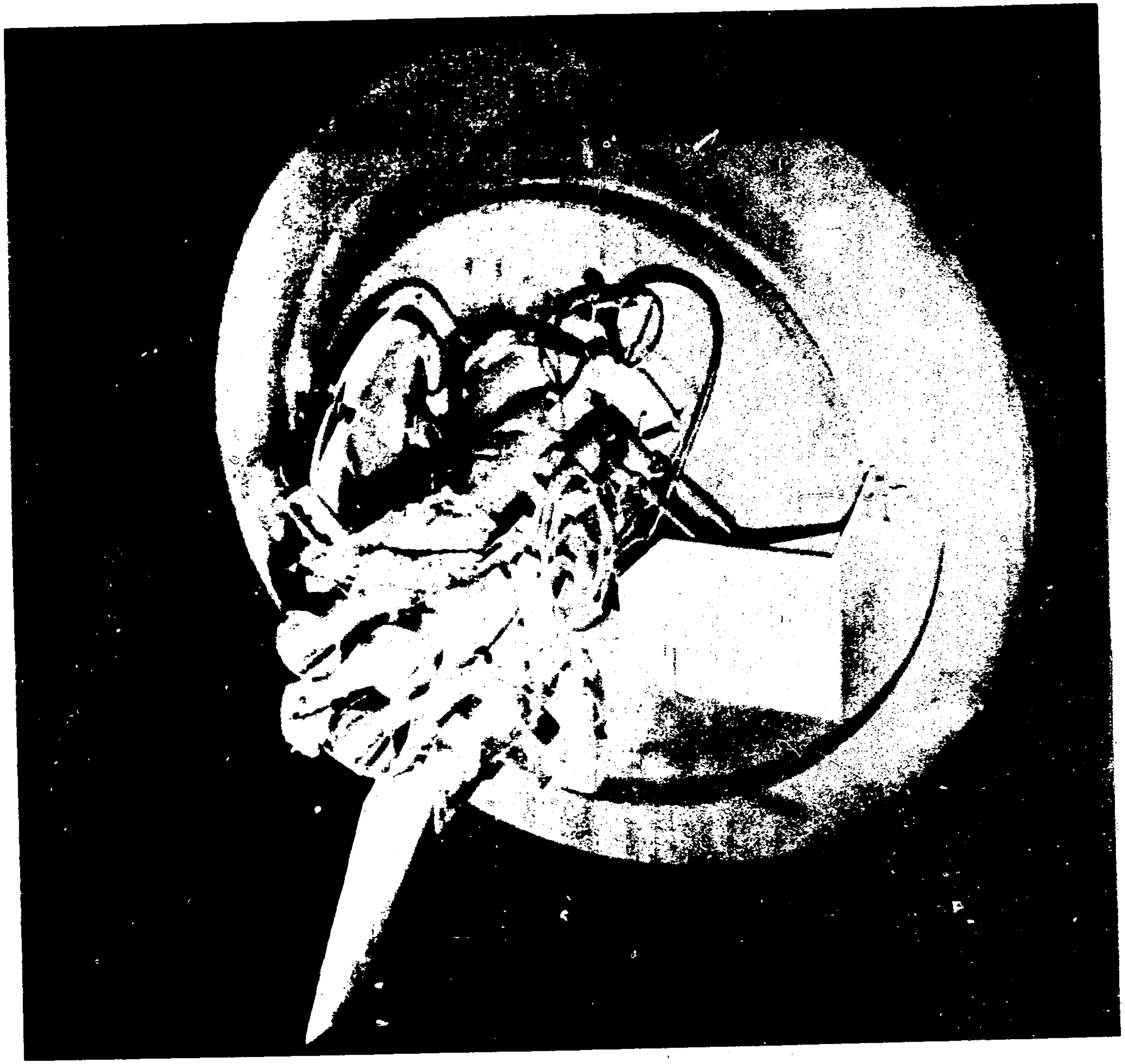


Figure 5.12. - Heating test of base heat shield at simulated high altitude; model no. 25 in the MSFC Impulse Base Flow Facility.

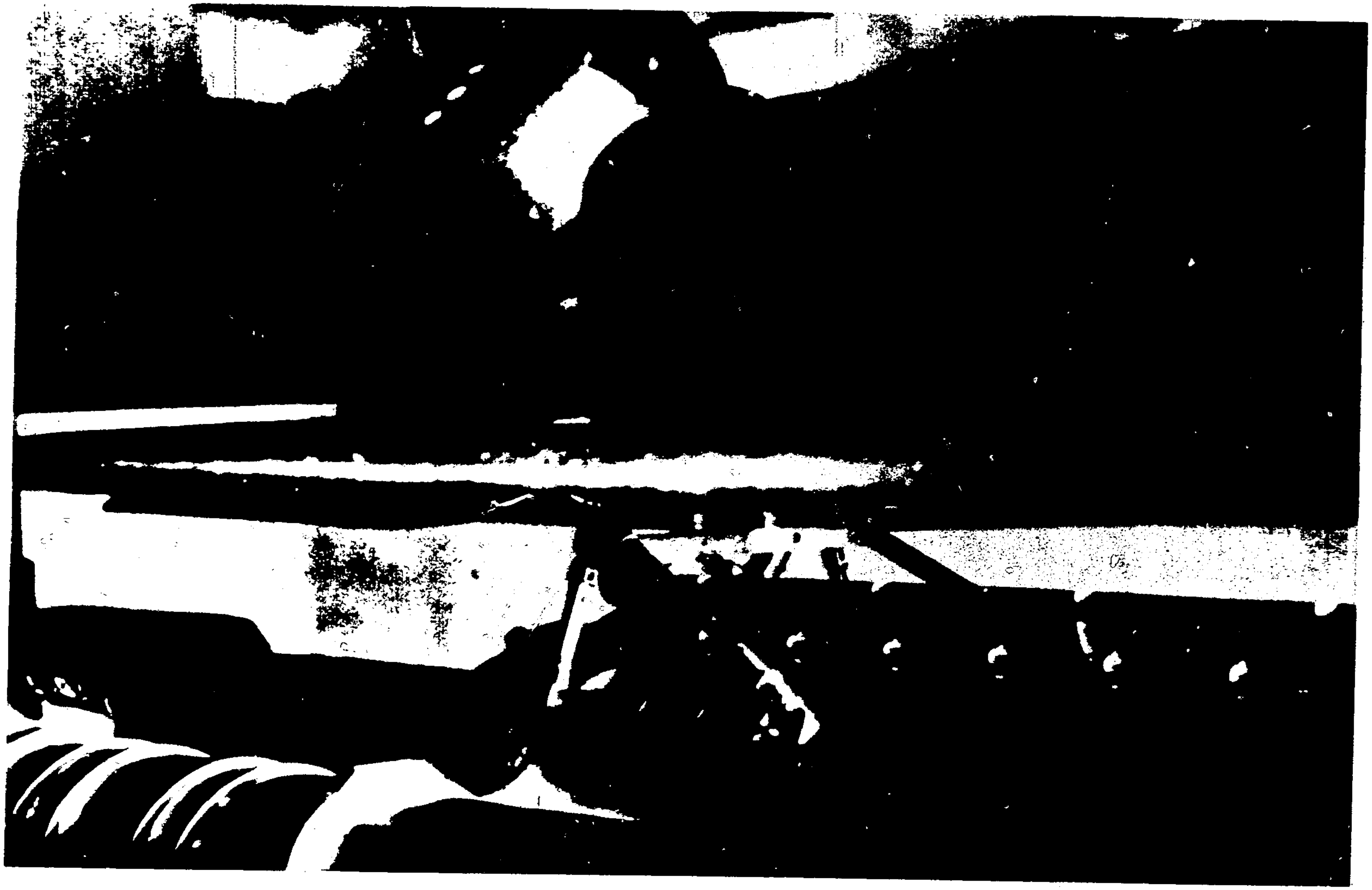


Figure 5.13. - Heating tests with high fidelity models; closeup of aft Orbiter/ET attach structure, model no. 60 in the AEDC-A Tunnel.

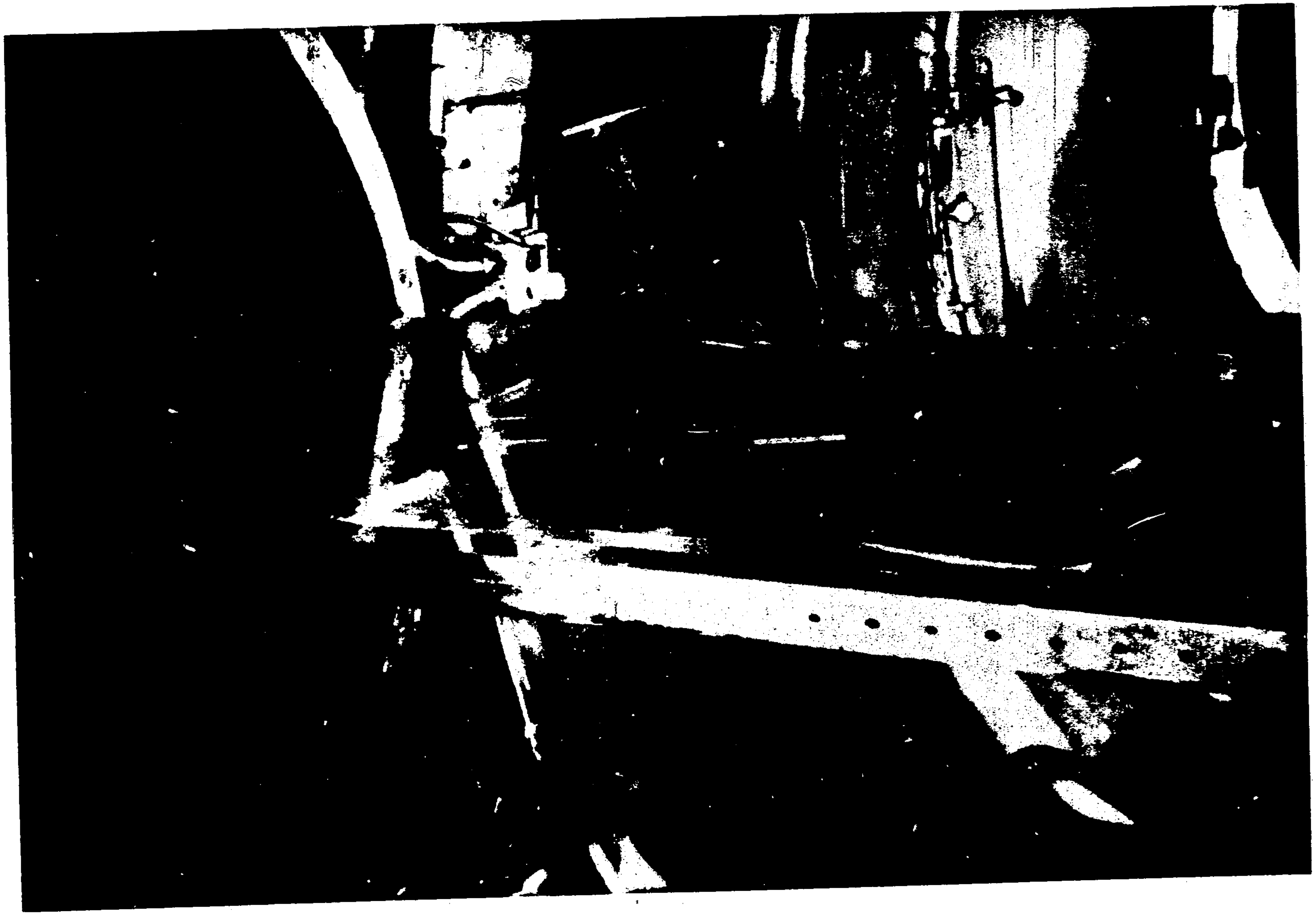
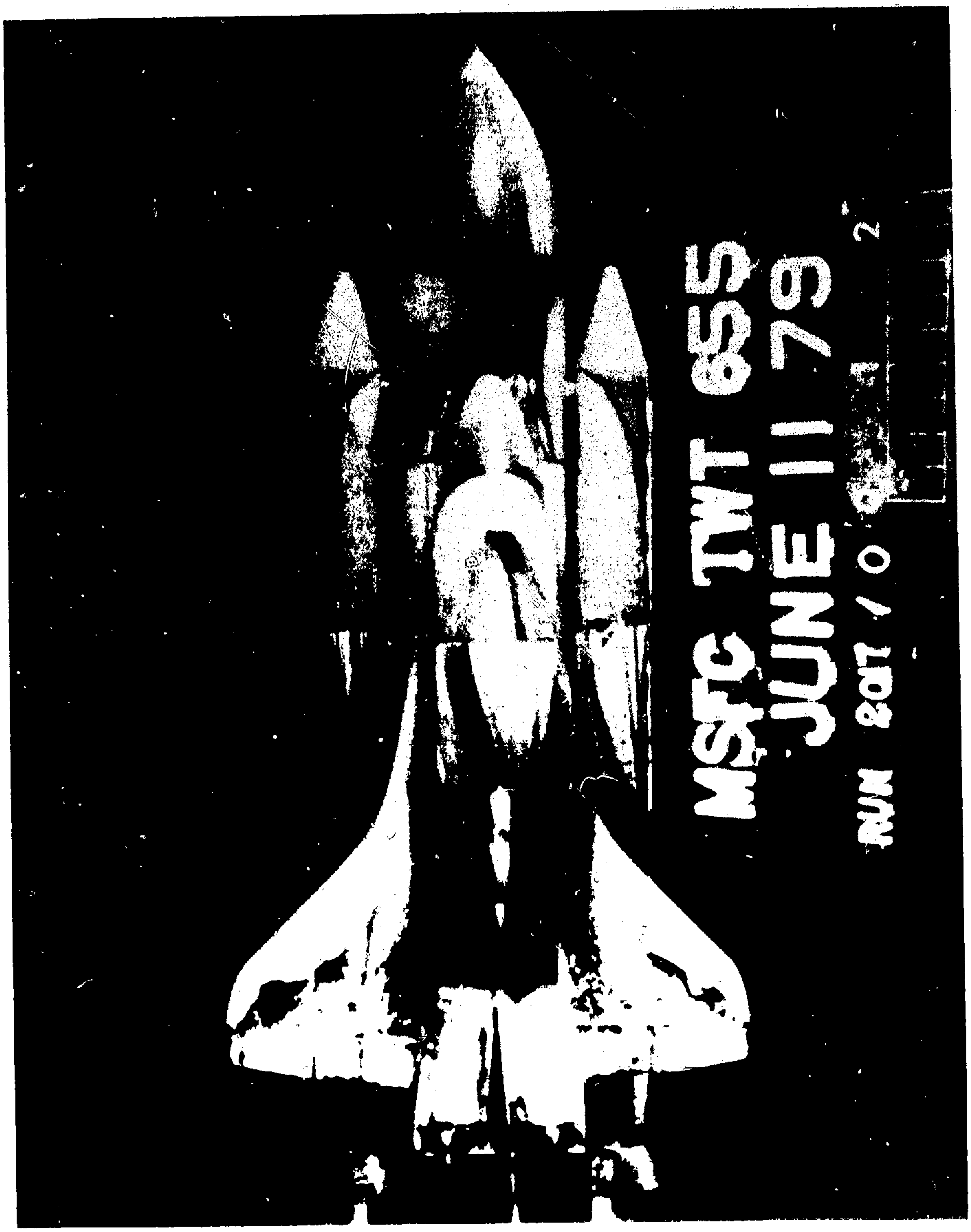
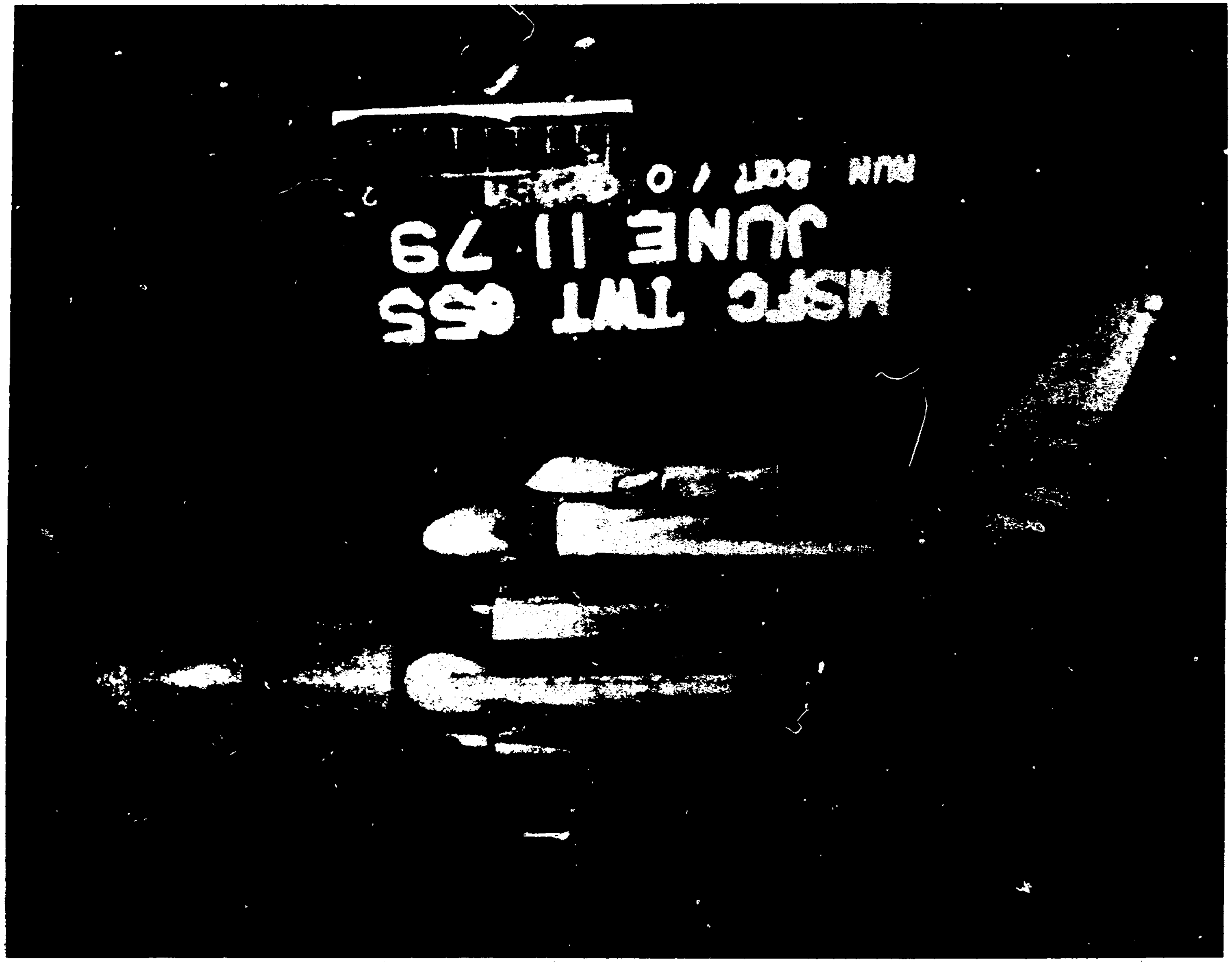


Figure 5.14. - Heating/pressure test of full-scale tiles; model no. 58 in the ARC 3.5-ft Hypersonic Tunnel.



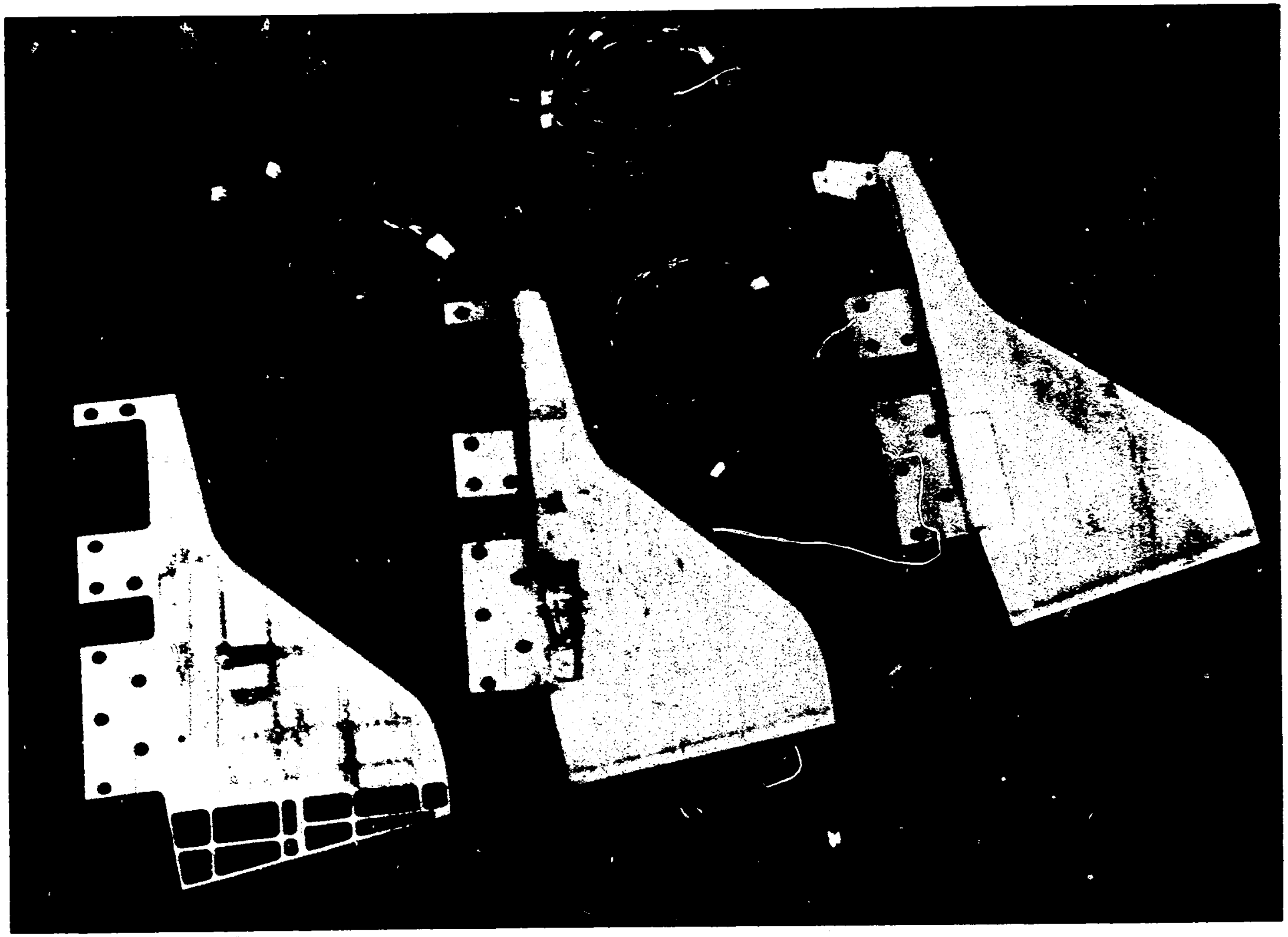
(a) Plan view, top.

Figure 5.15. - Oil flow test; model no. 74, MSFC 14-in. Trisonic Wind Tunnel.



(b) Side view, left.

Figure 5.15. - Concluded.



(a) Wind construction detail.

Figure 5.16. - Reflection plane model no. 30 tested in the LaRC 26-in. Transonic Blowdown Tunnel.

(b) Wing/body assembly.

Figure 5.16. - Concluded.

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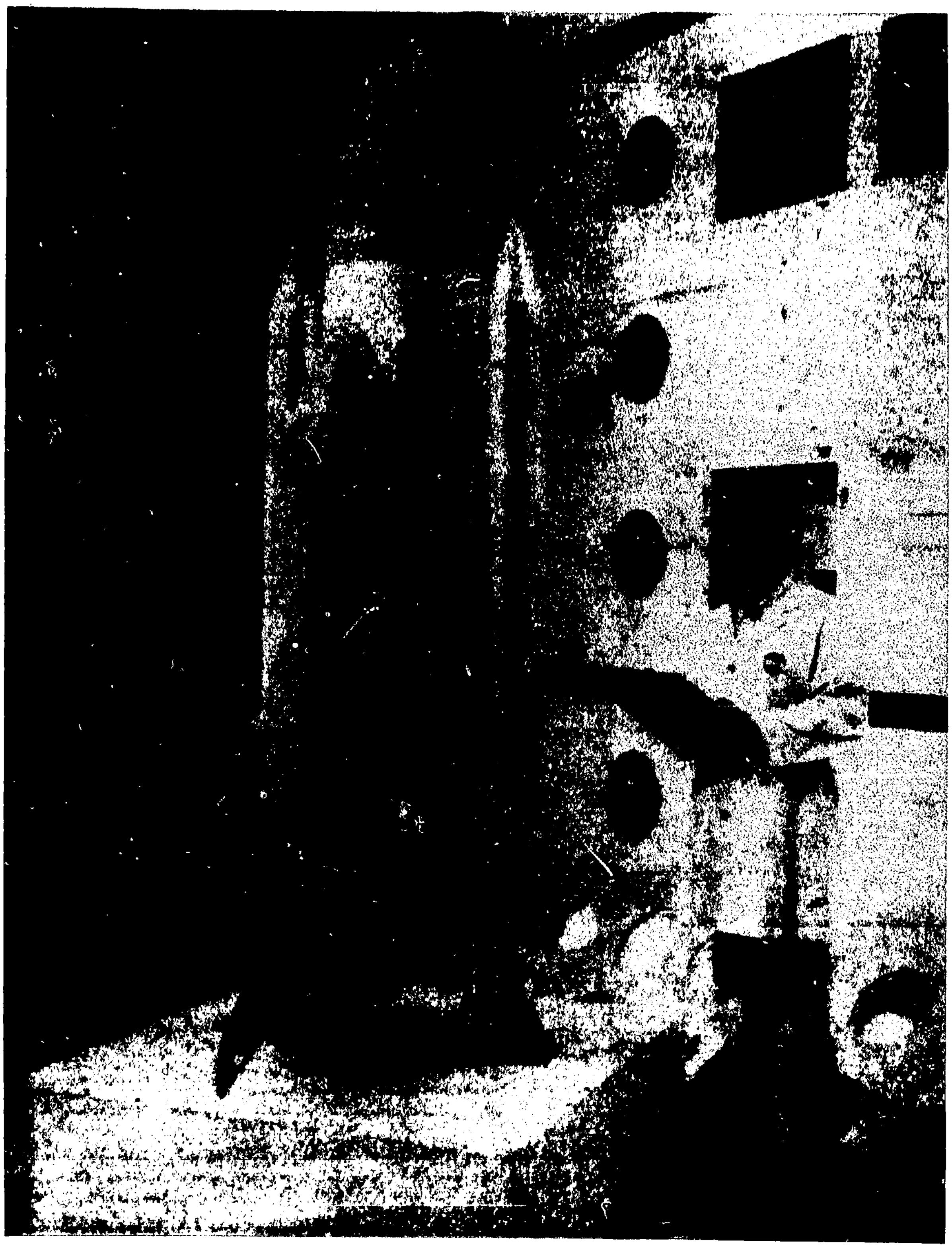
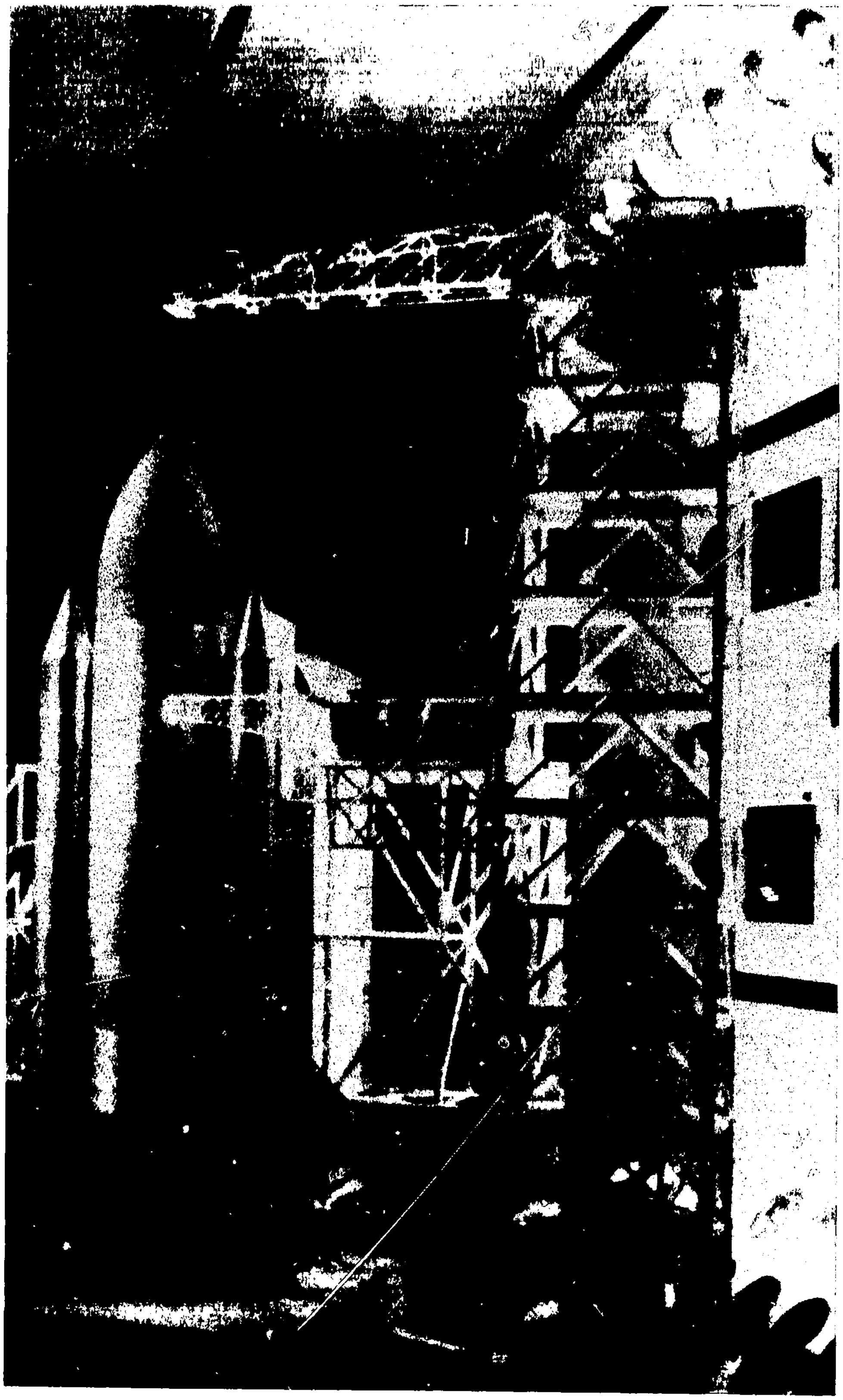


(a) Lift-off configuration.

Figure 5.17. - Ground winds effect tests in the LaRC 16-ft Transonic Dynamics Tunnel; model no. 100.



(b) Launch tower configuration.

Figure 5.17. - Concluded.

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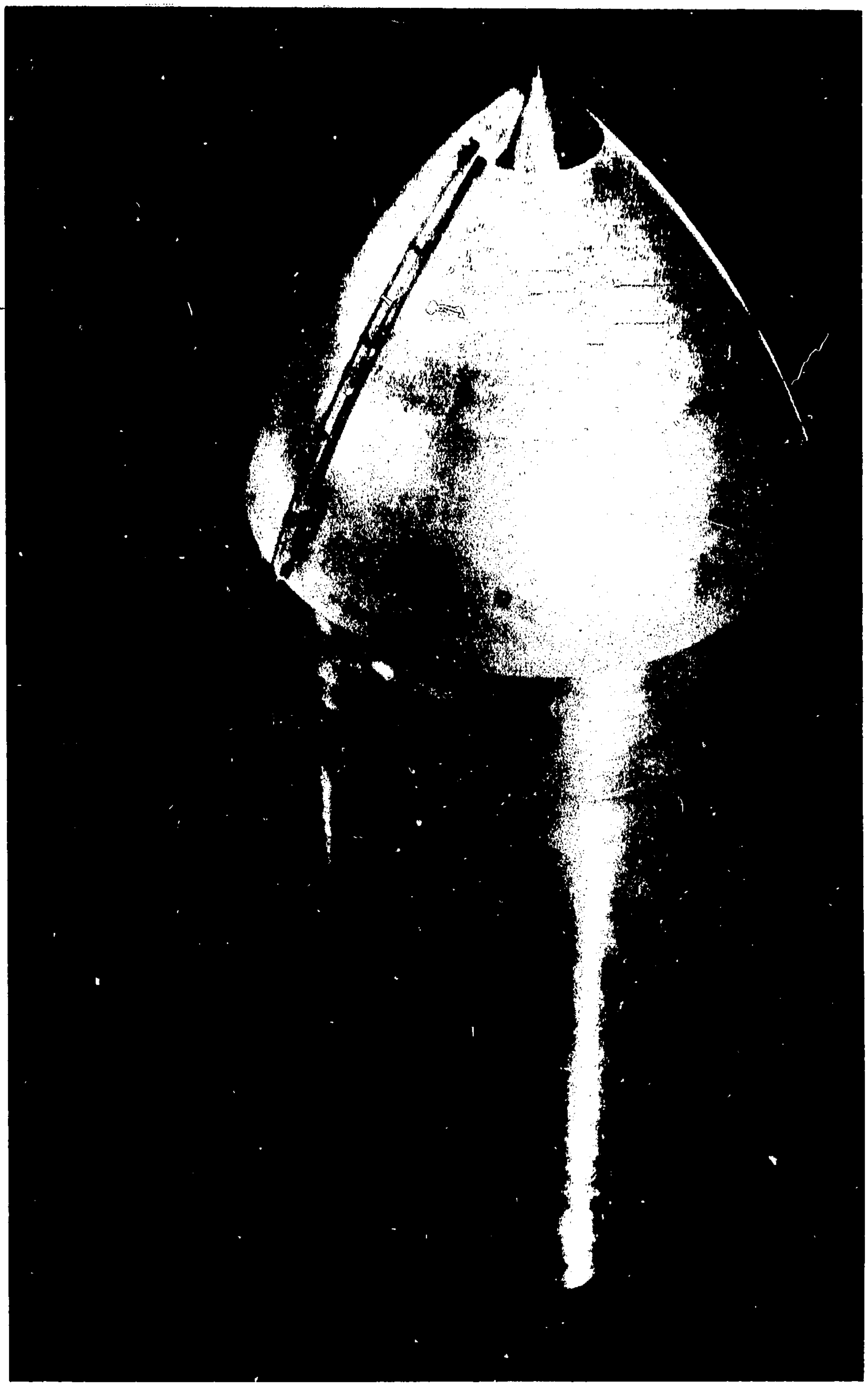

Figure 5.18. - AADS test in the ARC 14-ft Transonic Wind Tunnel; seven percent forebody model no. 68.

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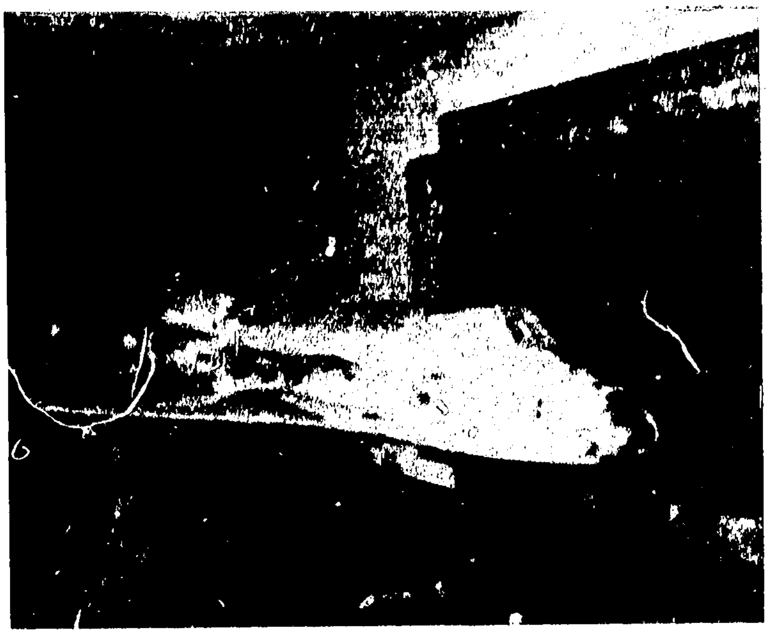
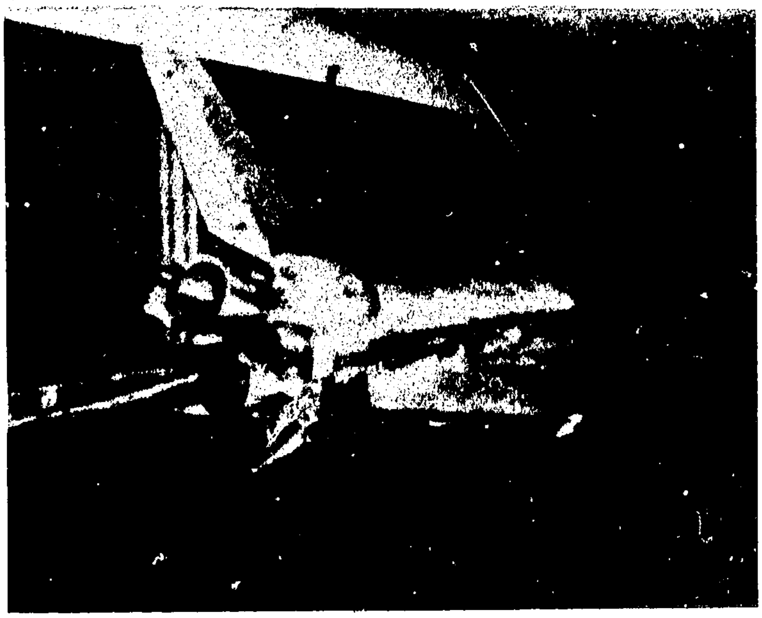


(a) Aft sting/strut floor mount; model no. 39 in the RI Low-Speed Wind Tunnel, front view.



(b) Aft sting/strut floor mount; model no. 39 in the RI Low-Speed Wind Tunnel, rear view.

Figure 5.19. - Orbiter test support arrangements.

(c) Aft sting/blade support strut; model no. 39 in the ARC 11-ft Transonic Wind Tunnel

Figure 5.19. - Continued.

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(d) Aft bent sting using roll to get α and β; model no. 49 in the AEDC-B Tunnel.

Figure 5.19. - Continued.

(e) Aft straight sting; model no. 106 in the AEDC-B Tunnel. Note the higher fidelity of this later model.

Figure 5.19. - Continued.

(f) Aft bent sting; model no. 106 at high α (AEDC-B).

Figure 5.19. - Continued.

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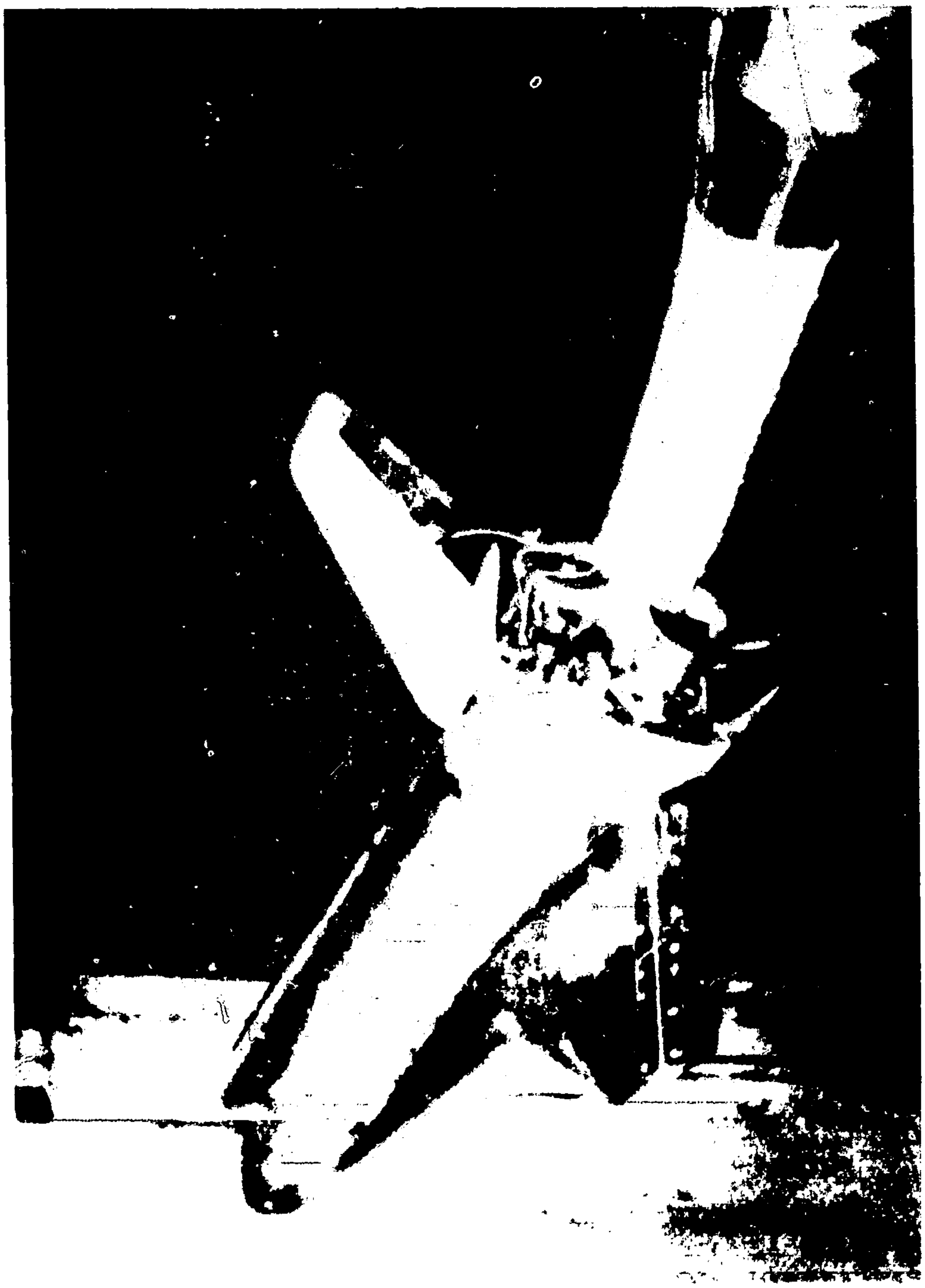
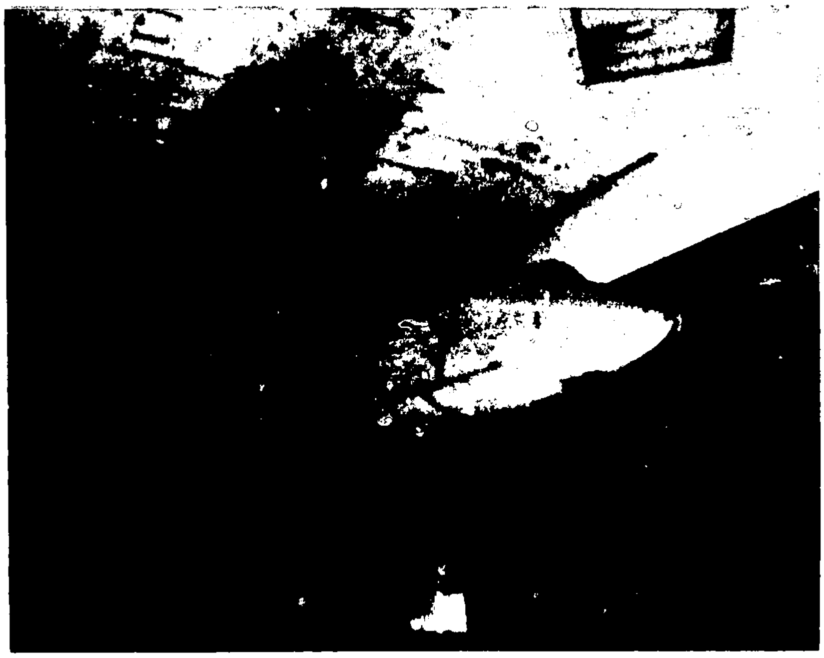


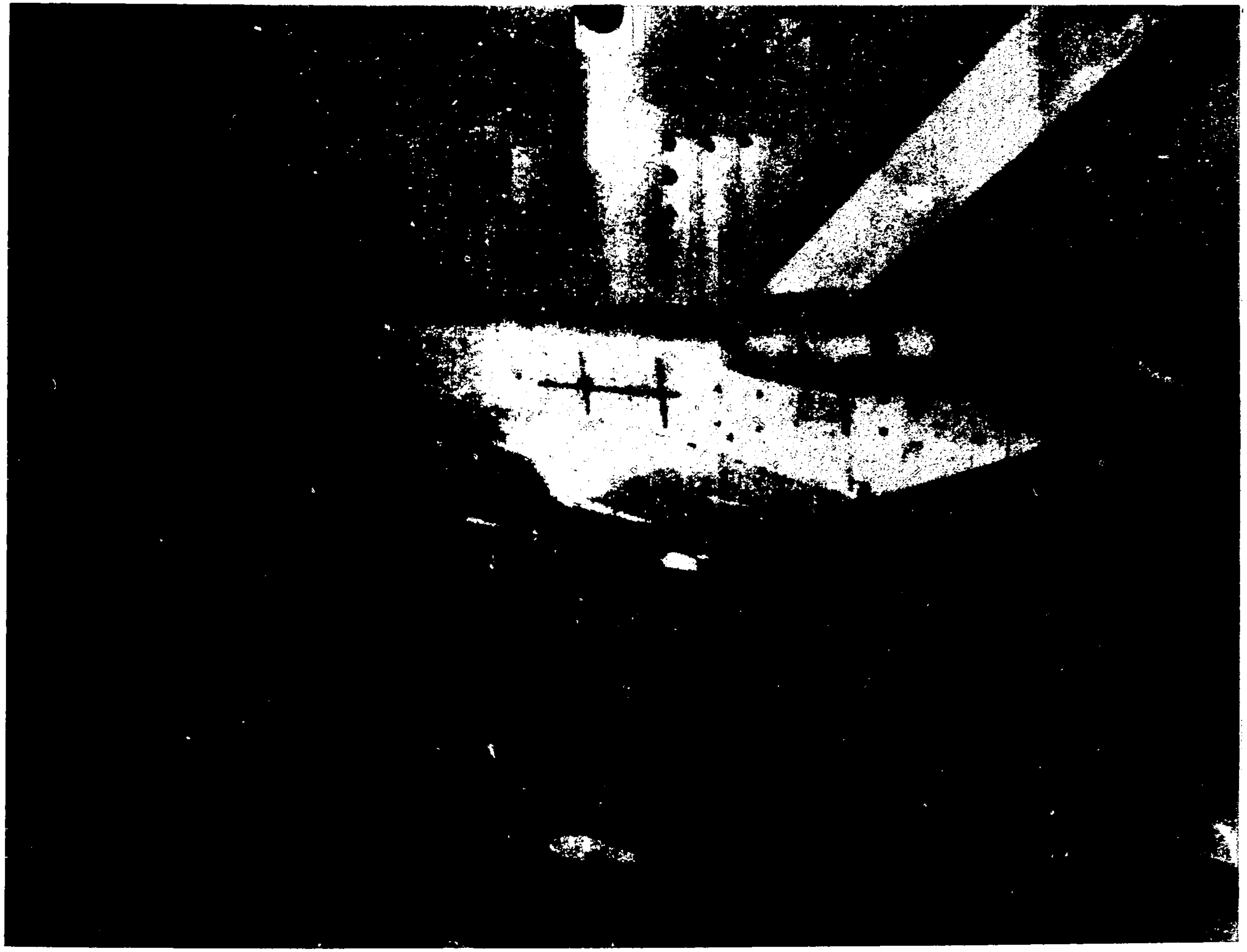
(g) Aft bent sting; model no. 105 at Low α (AEDC-A). Note the high fidelity model and the base pressure instrumentation.

Figure 5.19. - Continued.



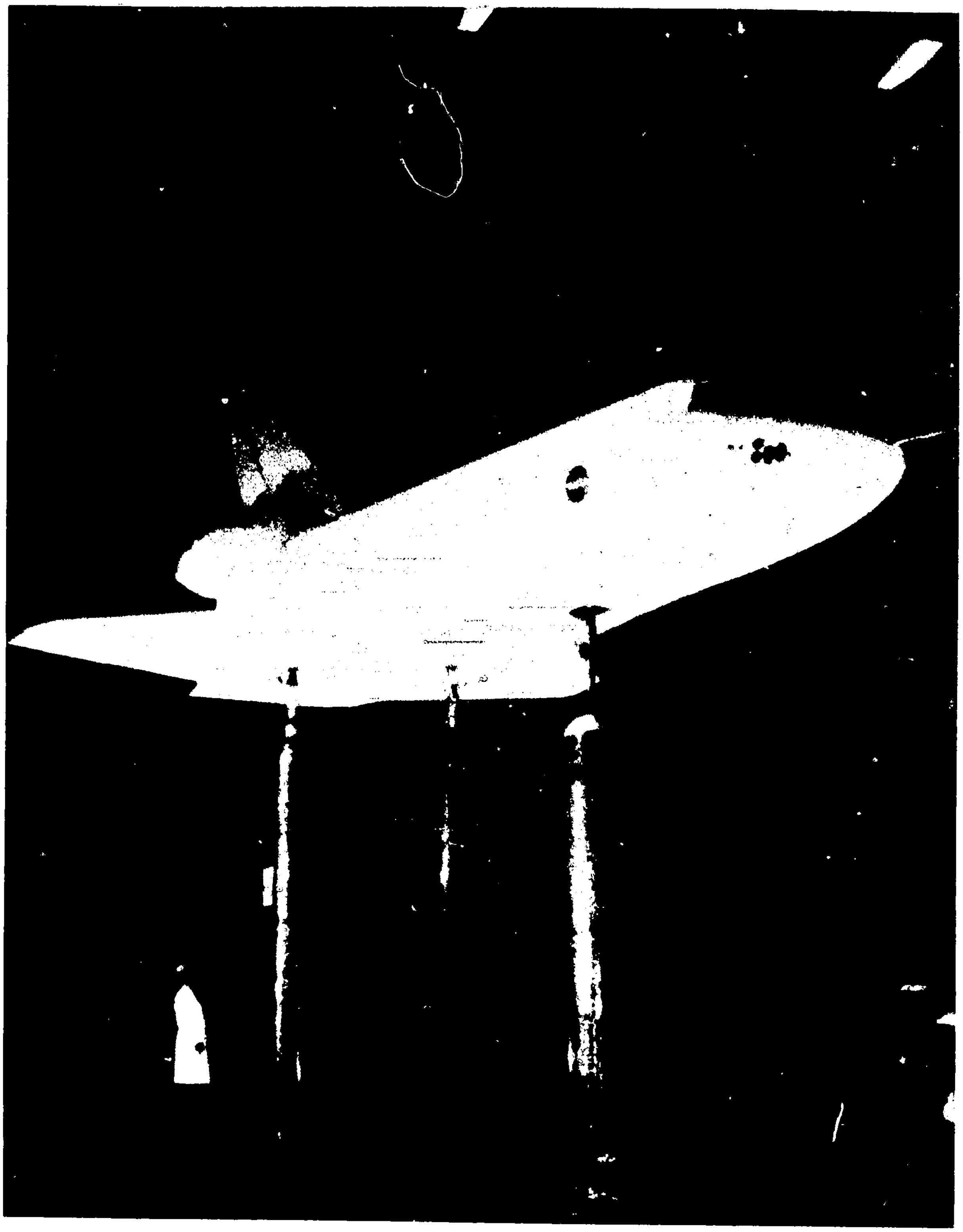
(h) Wing tip mount; model no. 43 in the RI Low-Speed Wind Tunnel.

Figure 5.19. - Continued.



(i) Three pylon strut support; model no. 45 in the ARC 12-ft Pressure Wind Tunnel.

Figure 5.19. - Concluded.



(a) Three pylon strut support; front view.

Figure 5.20. - Large scale model (36 percent) no. 76 in the ARC 40- by 80-ft Subsonic Wind Tunnel.

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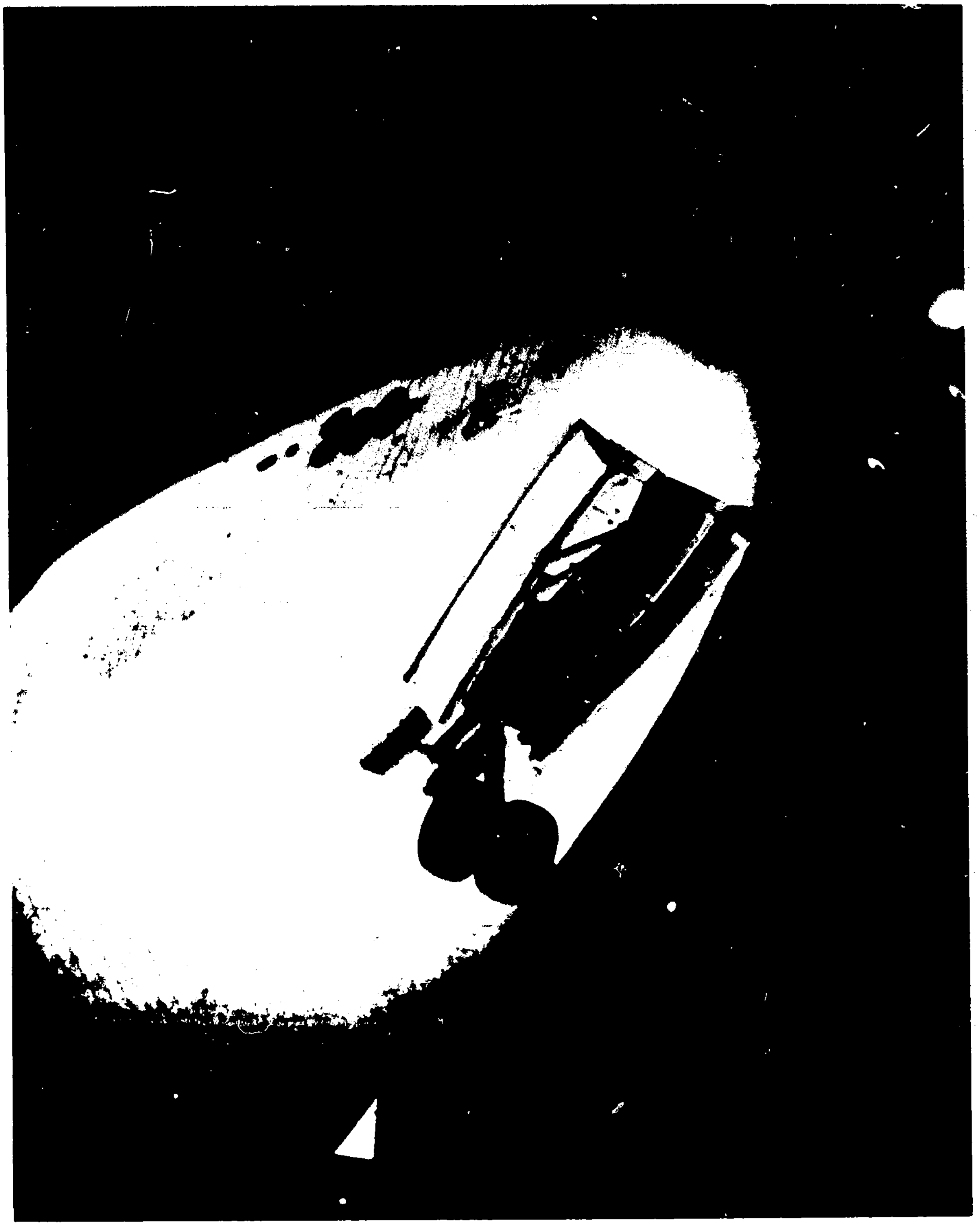

(b) Nose details showing tile simulation and the nose landing gear.

Figure 5.20. - Continued.

(c) Three pylon strut support; rear view.

Figure 5.20 - Continued.

(d) Tailcone (ALT) configuration.

Figure 5.20 - Concluded.

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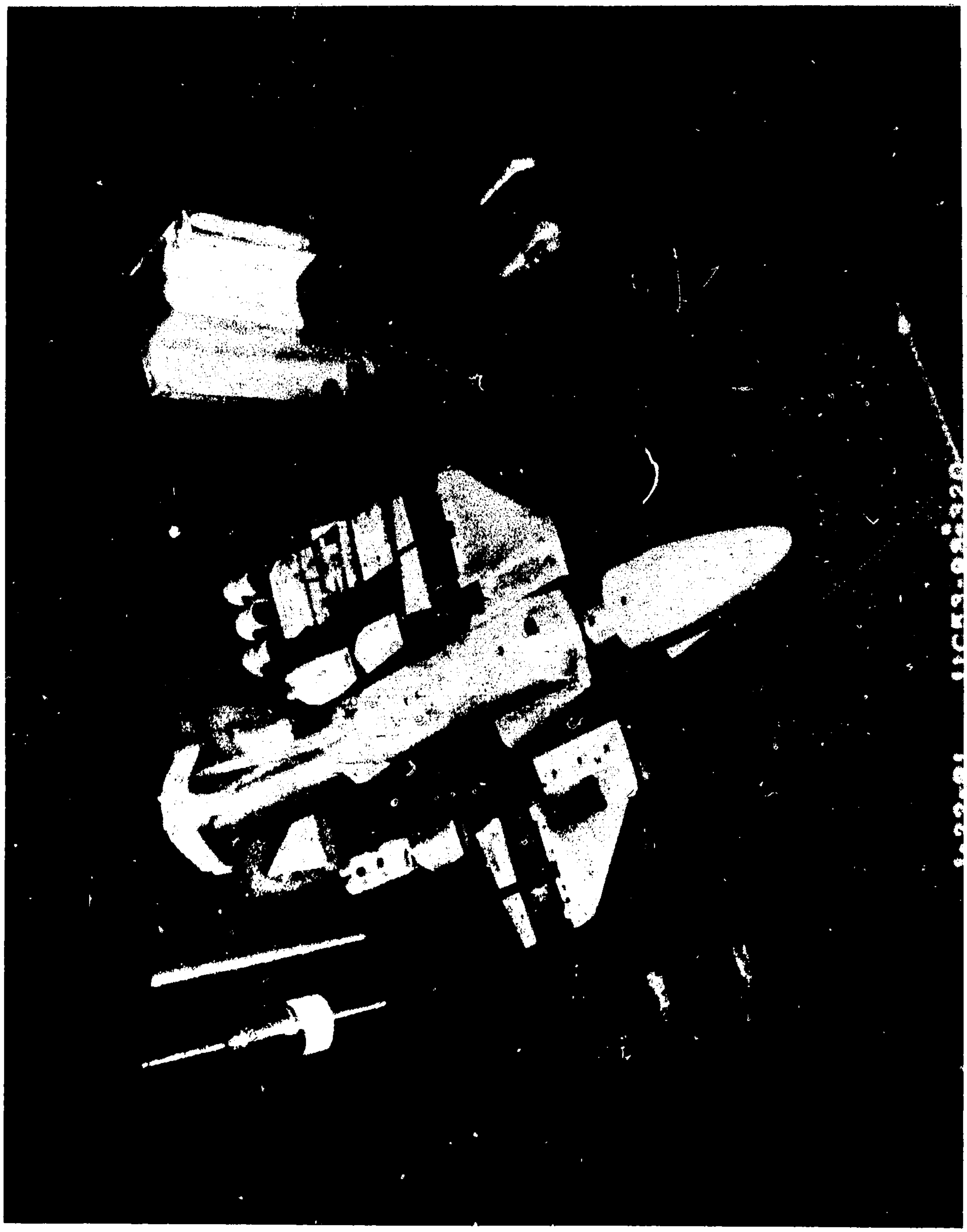
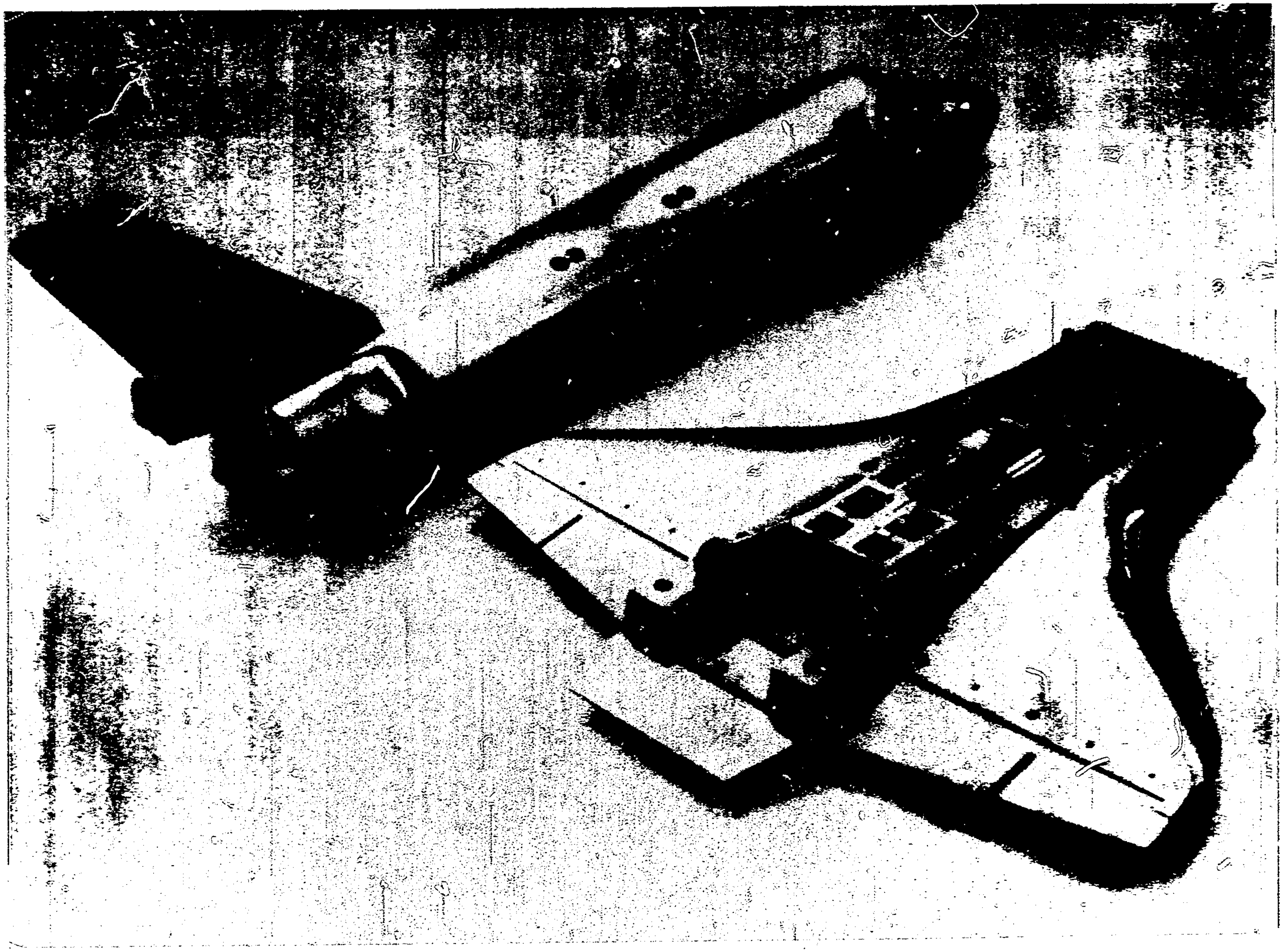


(a) Model no. 72; exploded view of bolt-on parts.

Figure 5.21. - Control surface deflection models.

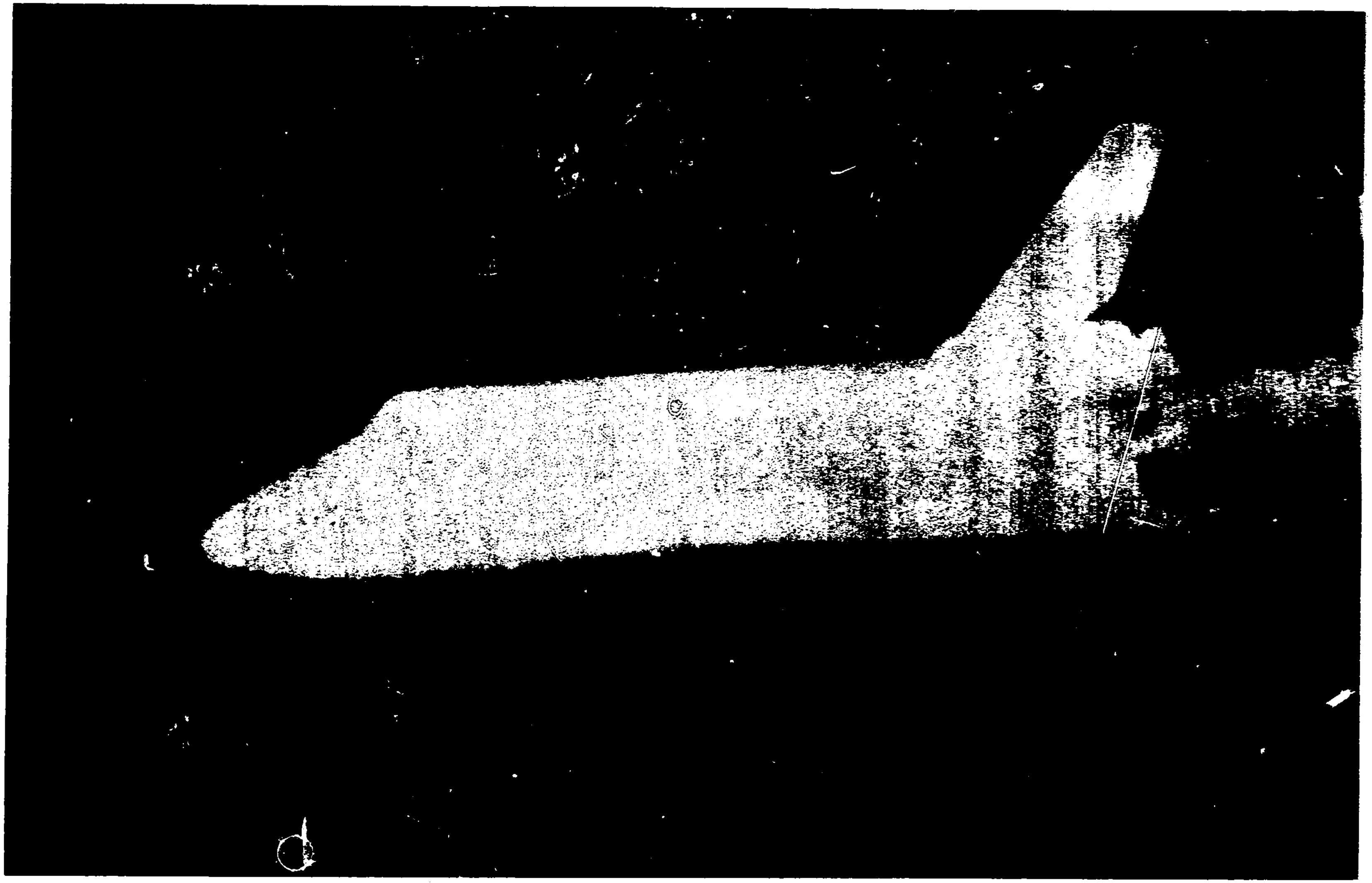
(b) Model no. 72; assembled view of bolt-on parts.

Figure 5.21. - Continued.



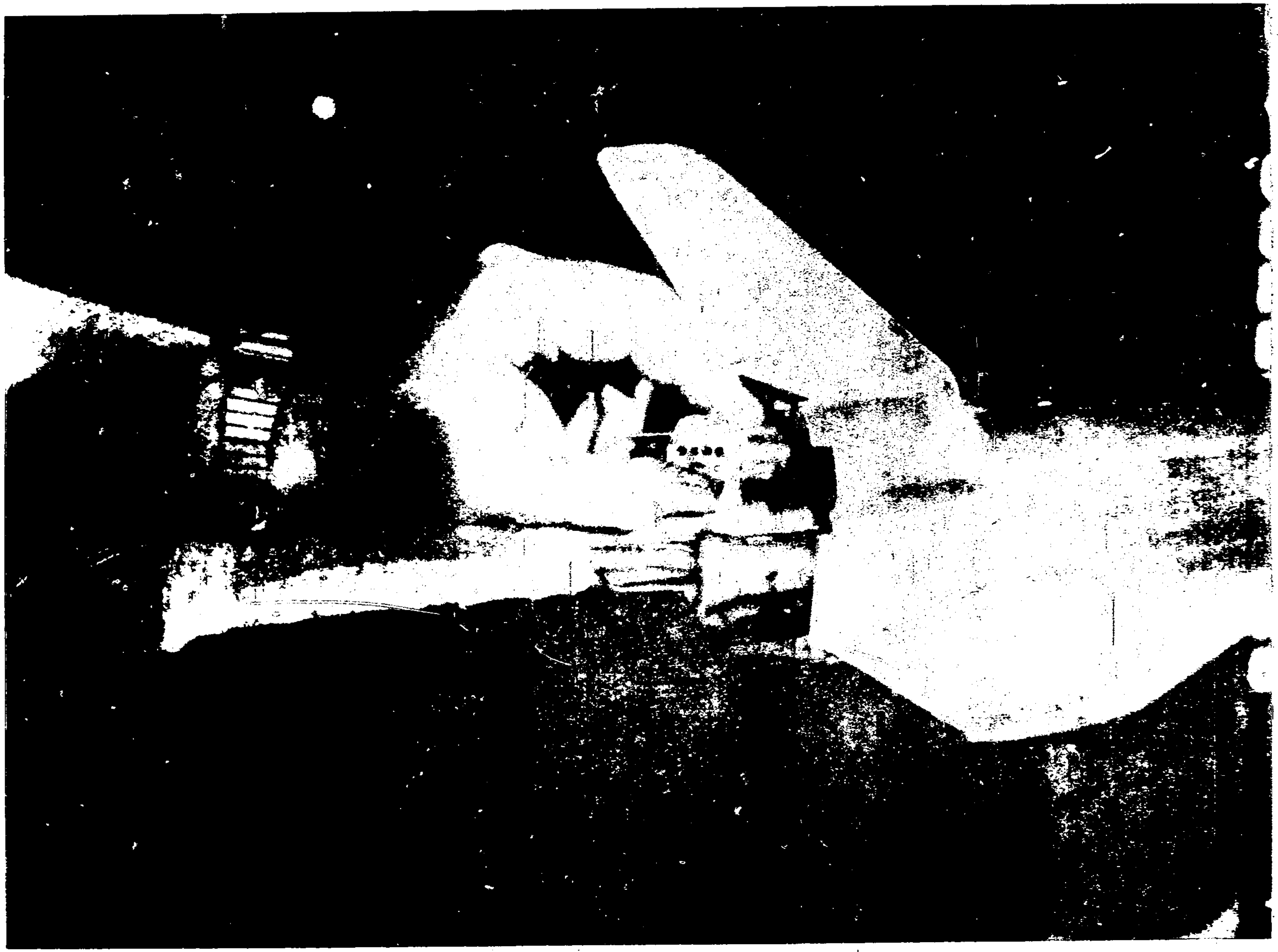
(c) Remote elevon model no. 44 showing internal mechanism.

Figure 5.21. - Continued.



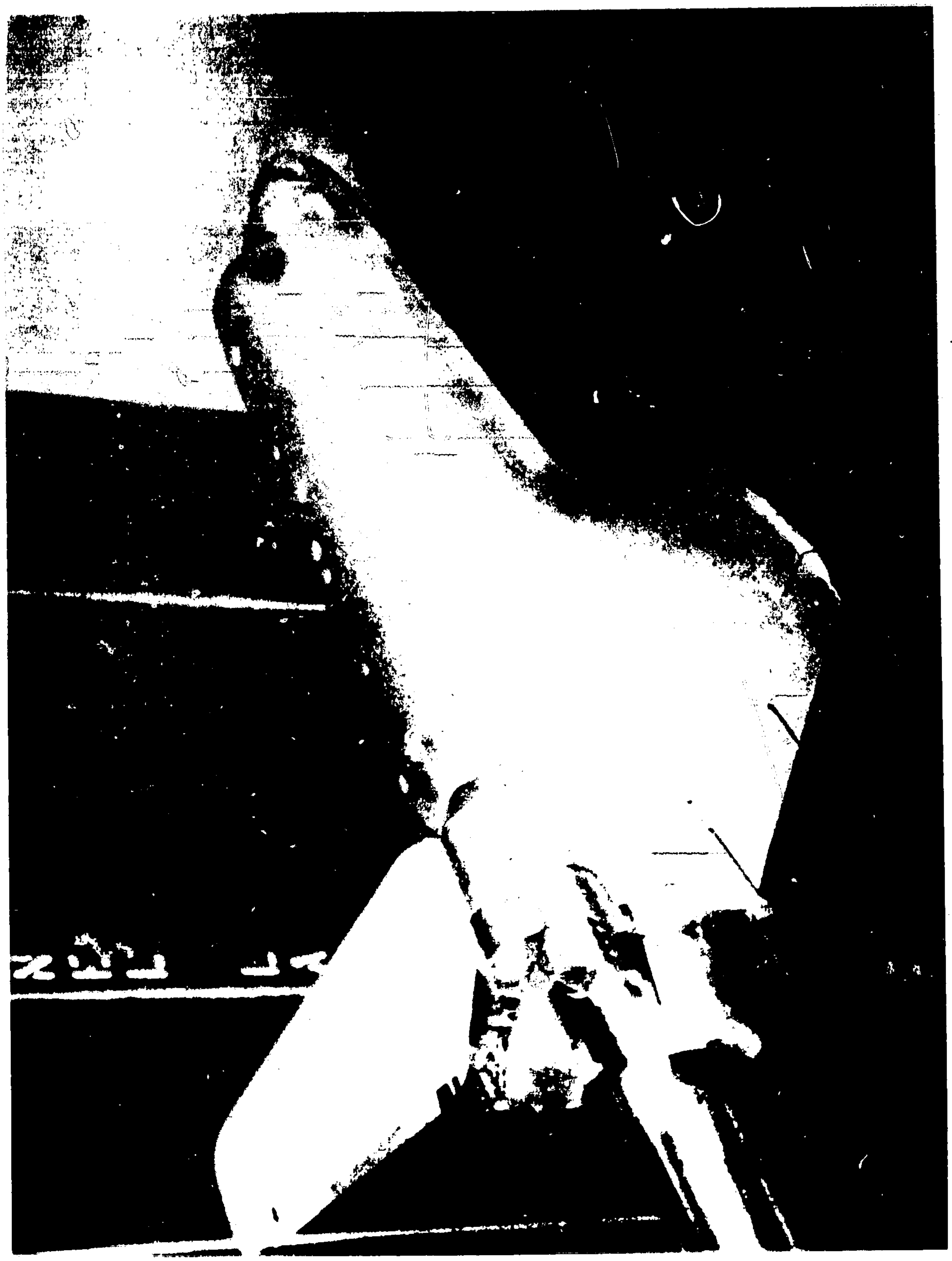
(d) Remote elevon/bodyflap/rudder high fidelity model (Vehicle 102 outer moldlines) no. 106.

Figure 5.21. - Concluded.



(a) RCS pod being installed.

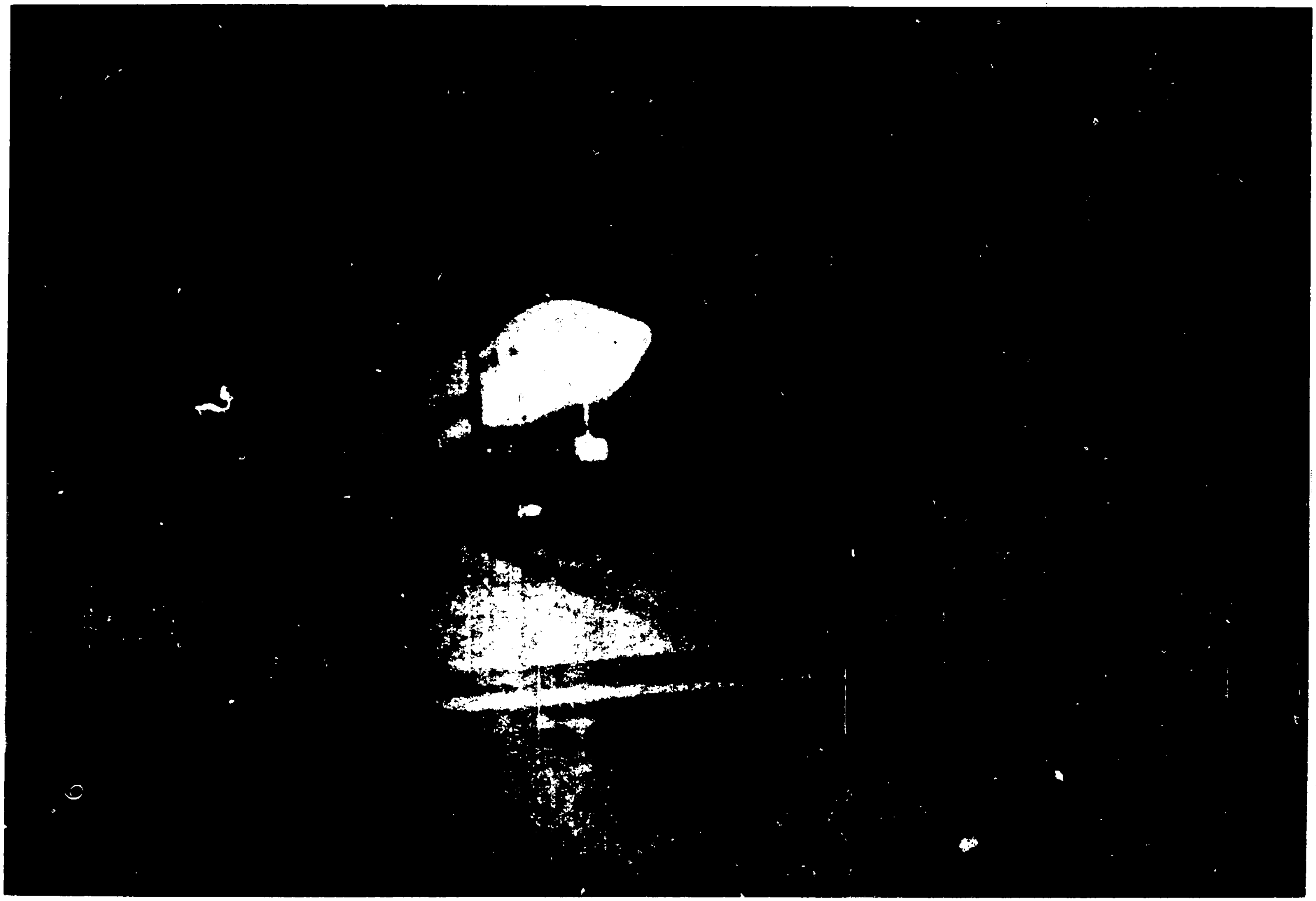
Figure 5.22. - RCS jet simulation model no. 70 in the LaRC Unitary Plan Wind Tunnel.



(b) Assembled model.

Figure 5.22. - Concluded.

Figure 5.23. - Aeroelastic Model. Picture not available.



(a) Front view.

Figure 5.24. - Ground effects tests; model no. 95 in the LTV 15- by 20-ft Low-Speed Wind Tunnel with moving ground plane.

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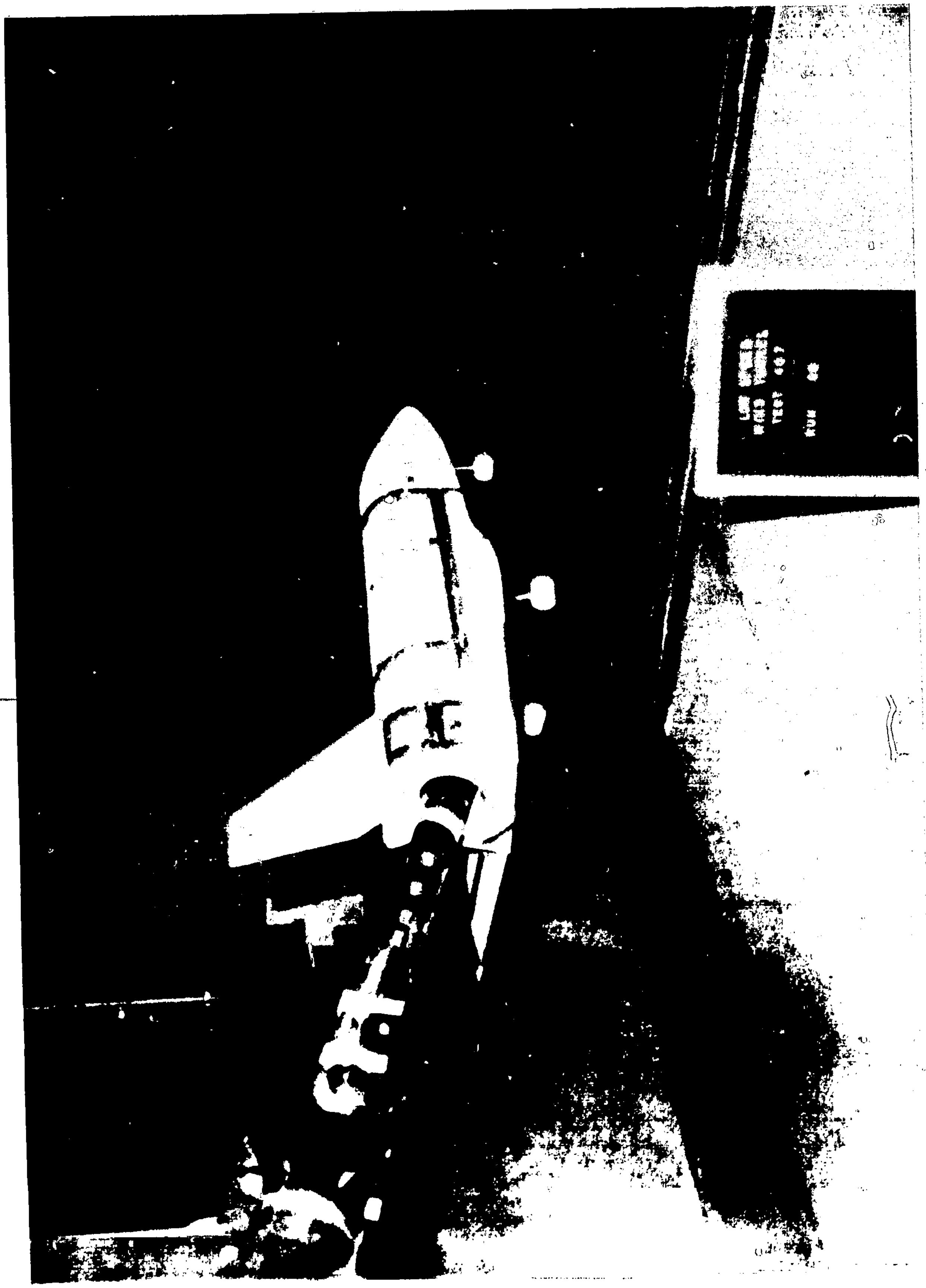

(b) Rear view.

Figure 5.24. - Concluded.

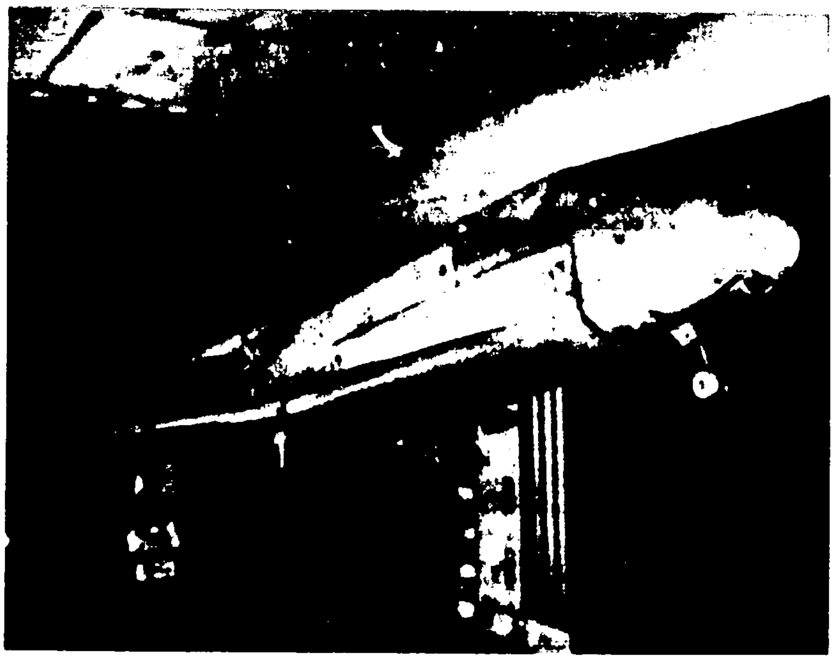

Figure 5.25. - Landing gear loads test; model no.16 in the RI Low-Speed Wind Tunnel.

(a) Side probes in the ARC 8- by 7-ft Supersonic Wind Tunnel.

Figure 5.26. - Orbiter ADS tests; forebody model no. 57.

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(b) Flight test boom (ALT) and side probes in the AEDC 16-ft Transonic Wind Tunnel.

Figure 5.26. - Concluded.

(a) Model no. 46 (for LaRC Variable Density Tunnel Test).

Figure 5.27. - Phase change paint models.

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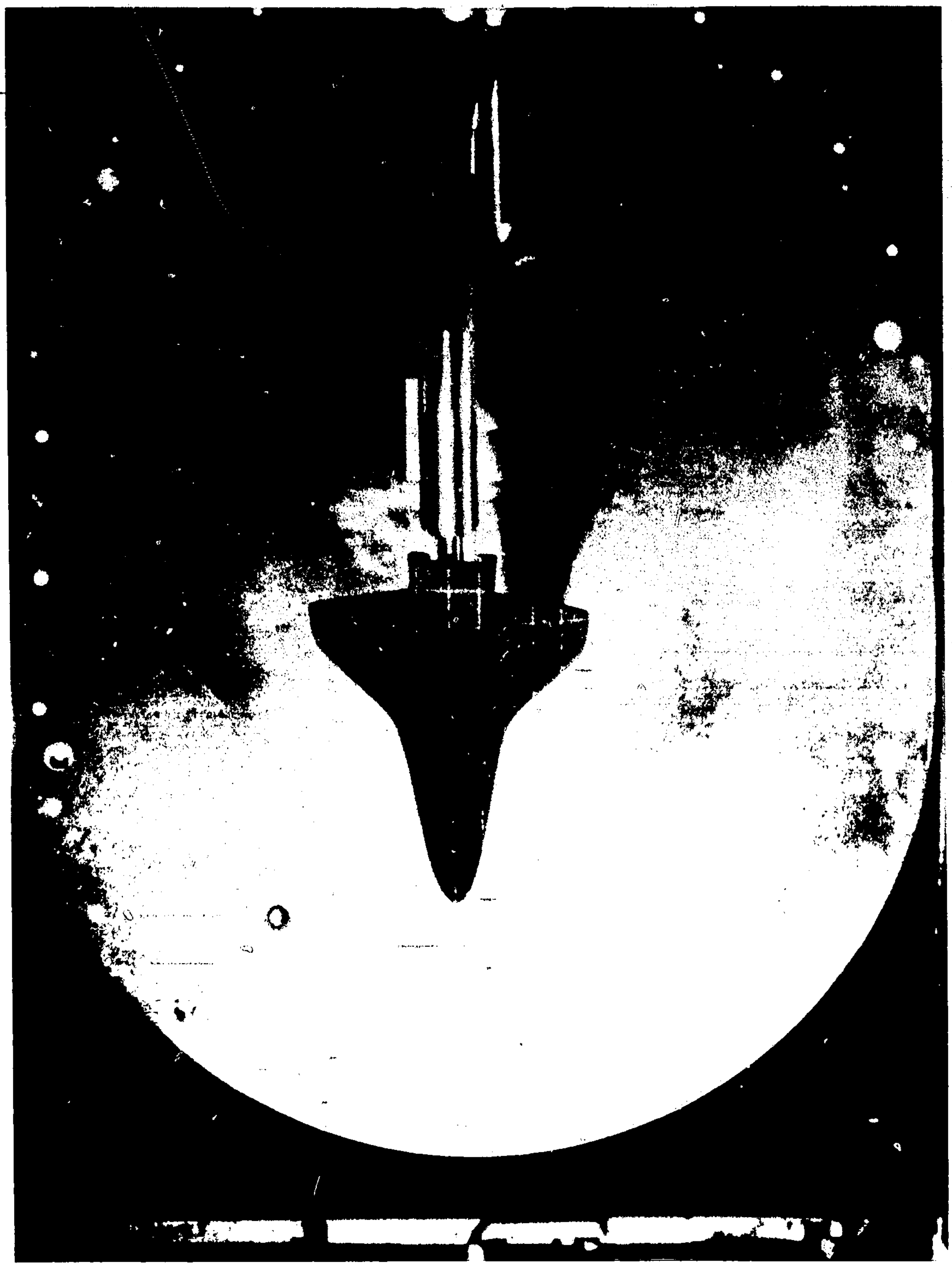


(b) Model no. 64 installed in the LaRC 31-in. Continuous Flow Hypersonic Tunnel.

Figure 5.27. - Concluded.

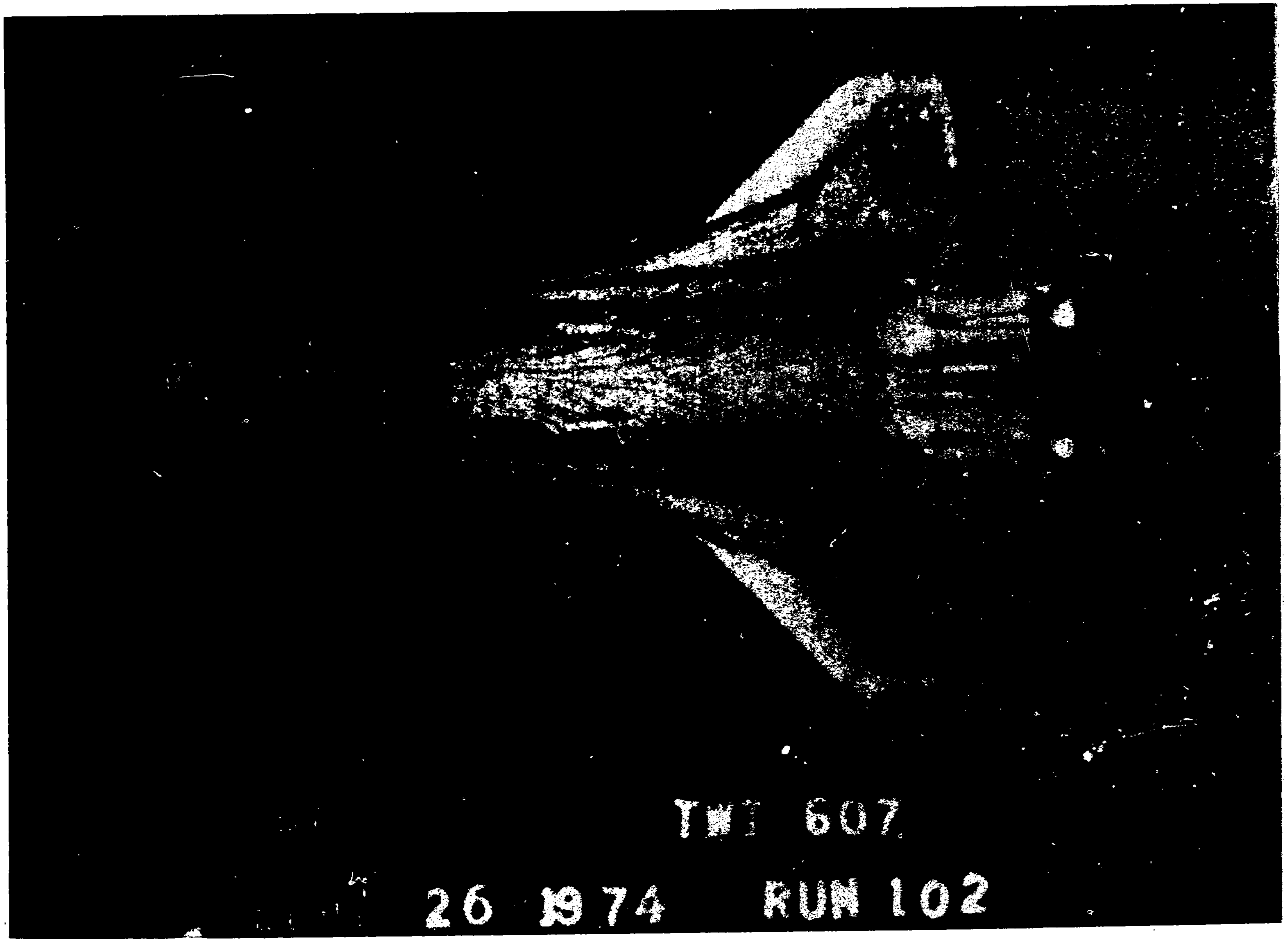
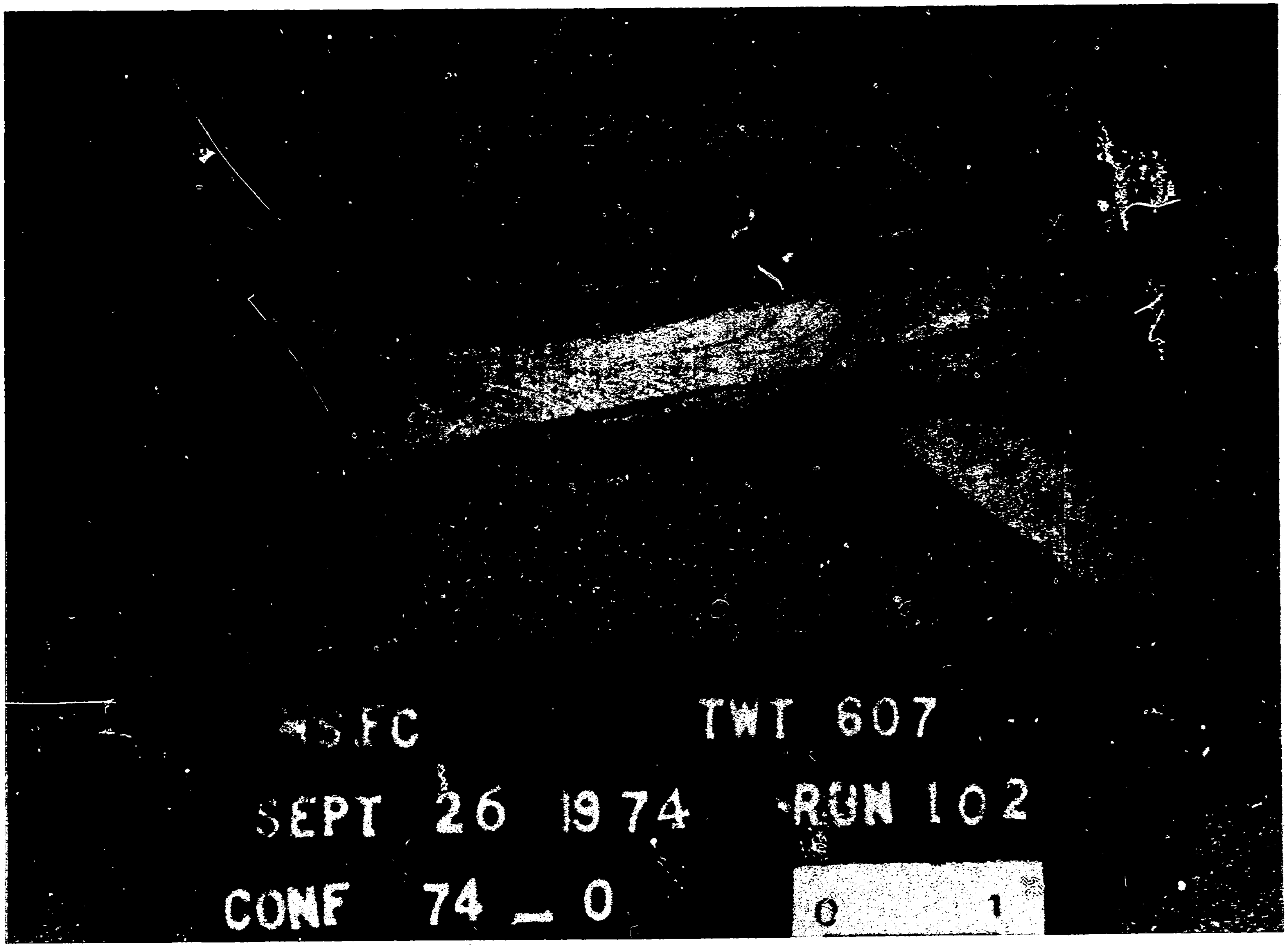


Figure 5.28. - Oil flow photographs; model no. 74 tested in the MSFC 14-in. Trisonic Wind Tunnel.



(b) Side view.

Figure 5.28. - Concluded.

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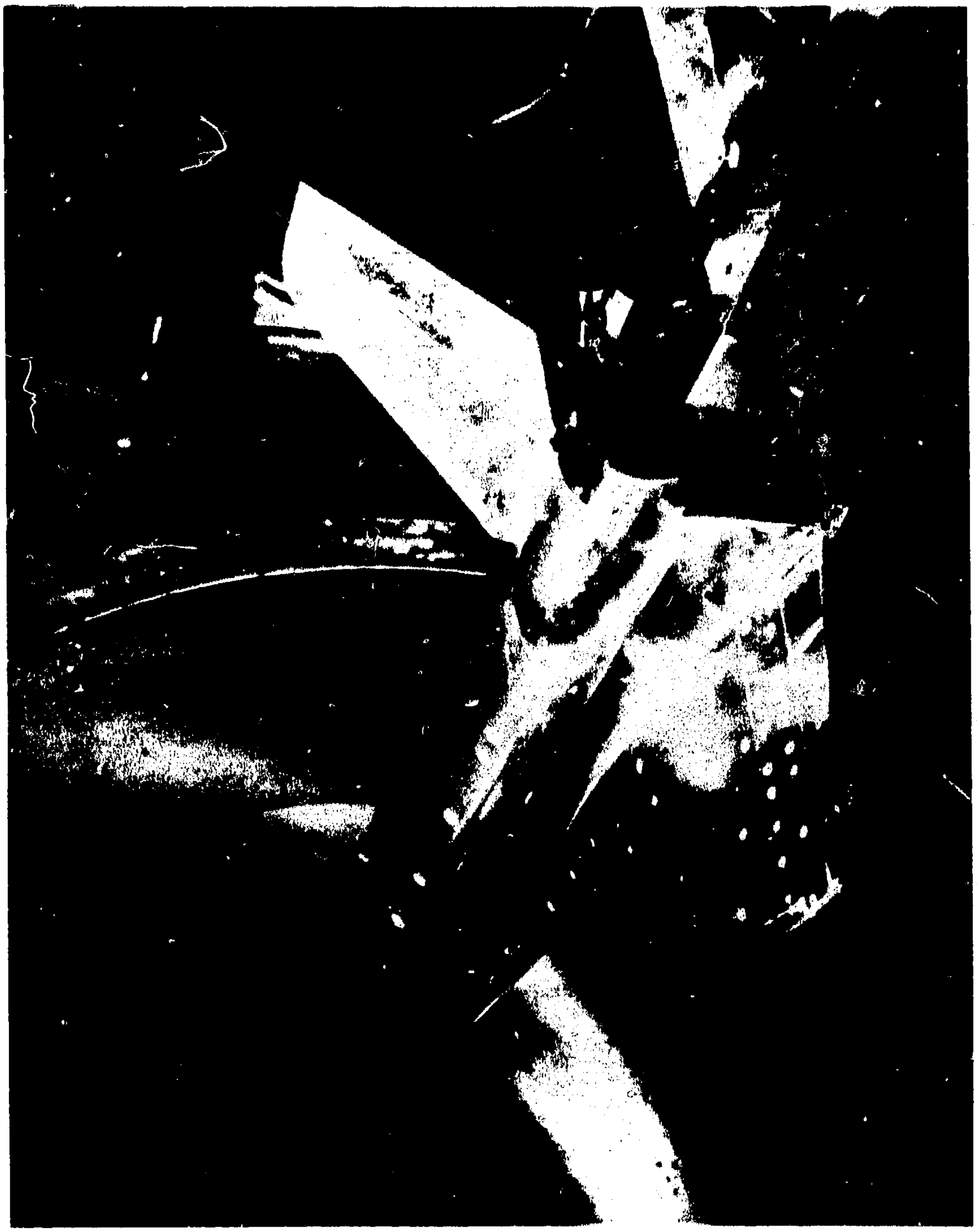
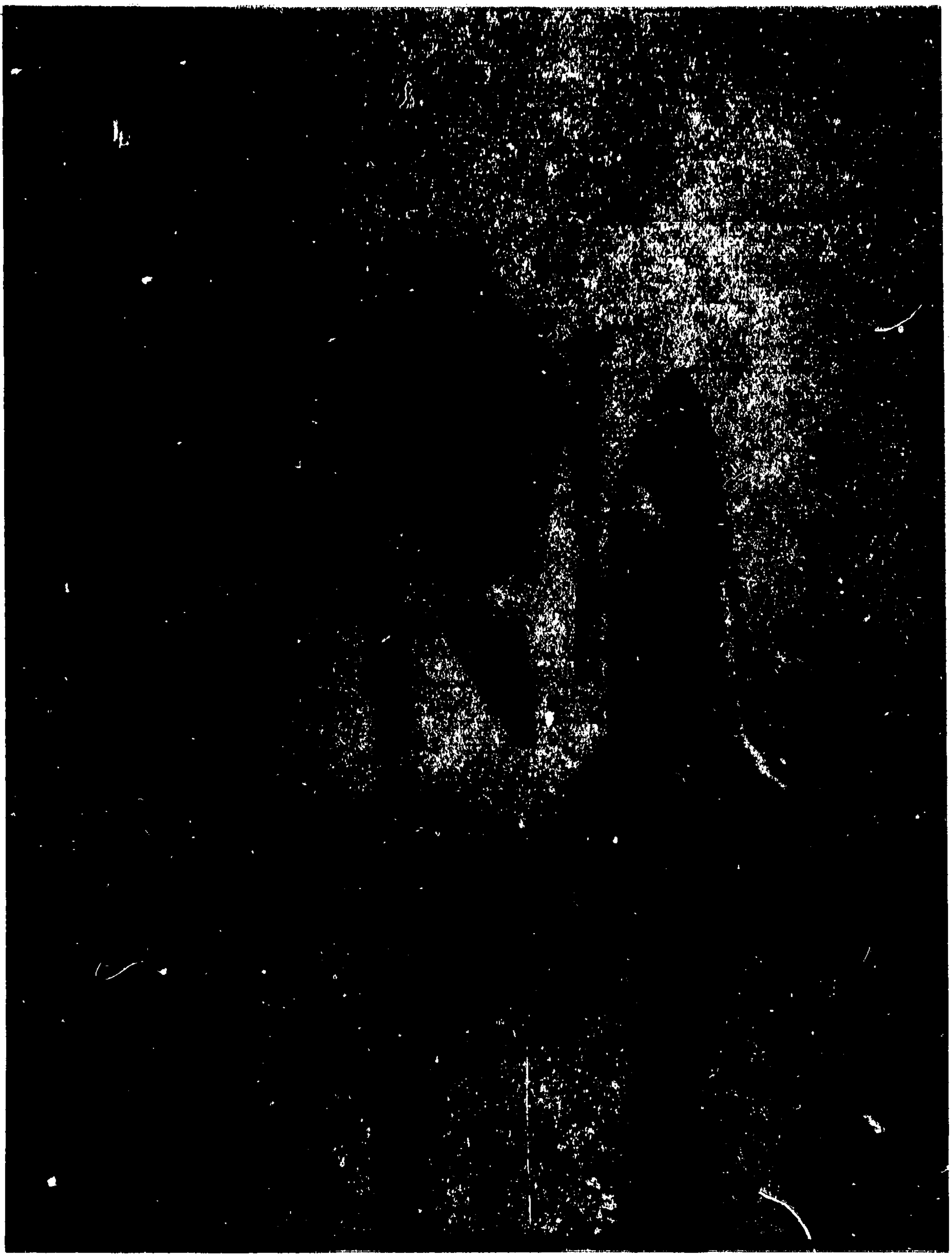


(a) Model no. 22 installed in the ARC 3.5-ft Hypersonic Tunnel.

Figure 5.29. - Orbiter thermocouple models.



(b) Model no. 50 showing thermocouple locations.

Figure 5.29. - Continued.

(c) Model no. 83 (forebody) installed in the AEDC-B Tunnel.

Figure 5.29. - Concluded.

Figure 5.30. - Heating test model no. 29 with probe for tunnel conditions in the AEDC-F Tunnel.

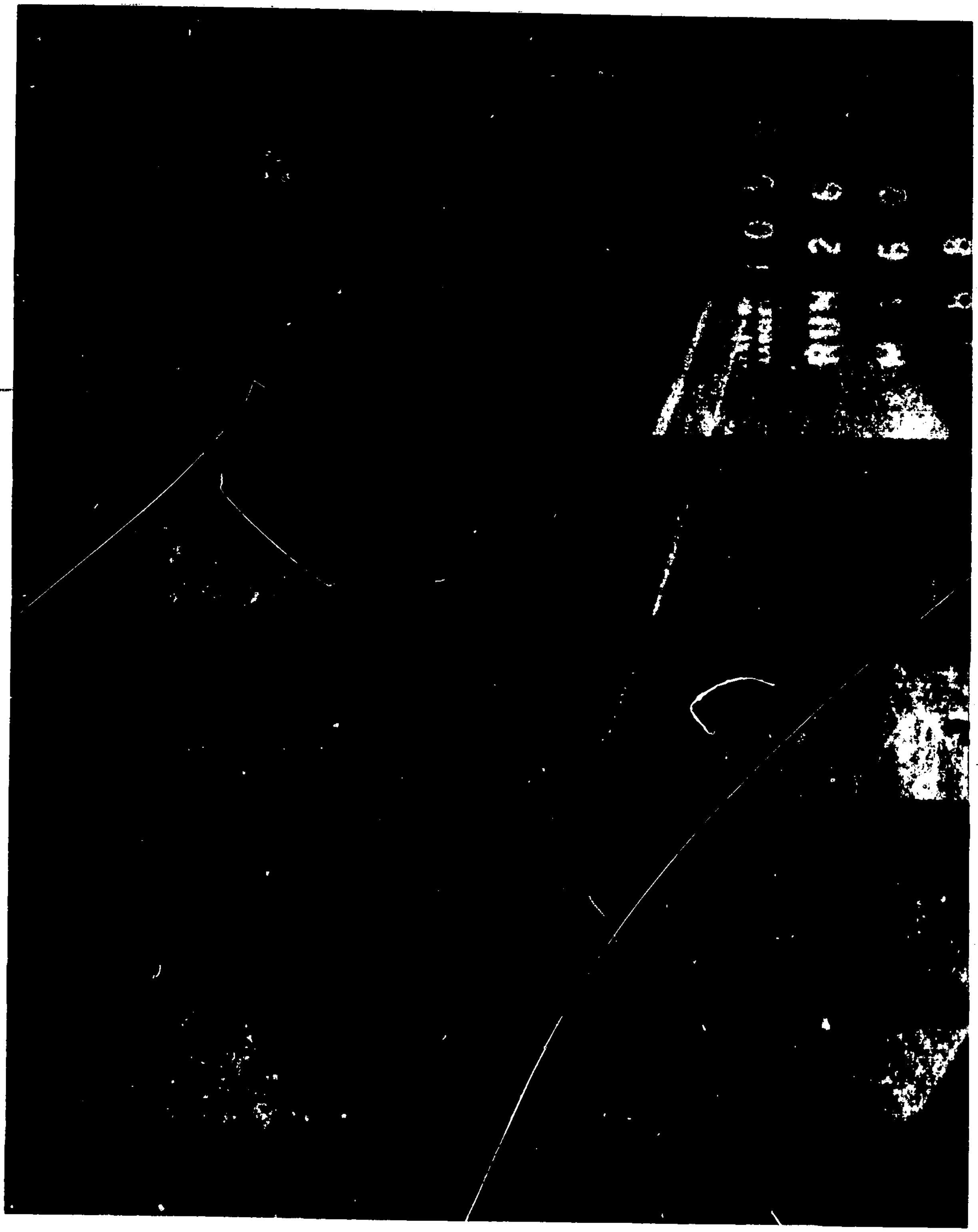
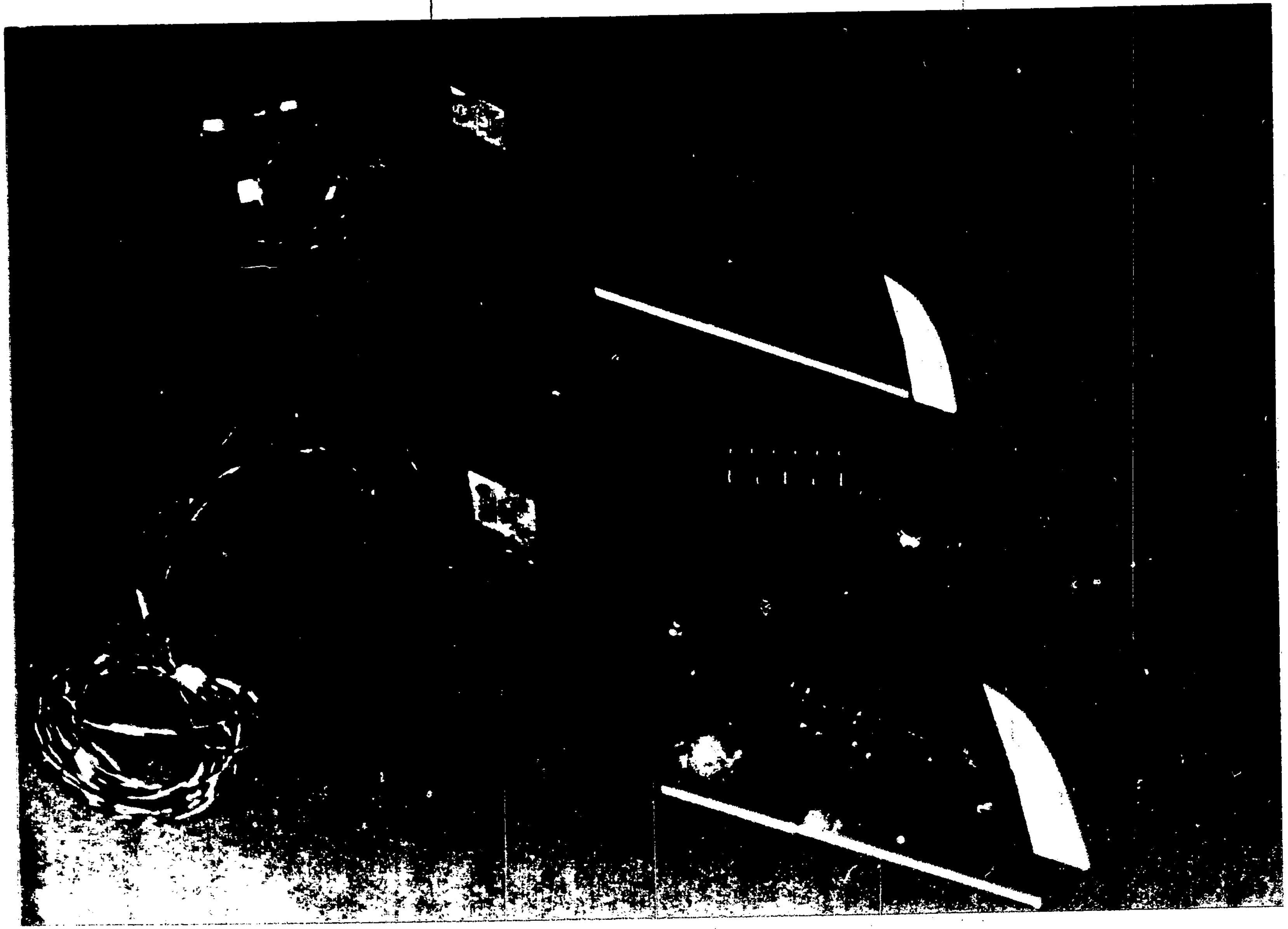


Figure 5.31. - Schlieren photo of model no. 26 in the LaRC Unitary Plan Wind Tunnel.



(a) Tail instrumentation.

Figure 5.32. - Flutter test of the vertical tail model no. 24 in the LaRC 26-in. Transonic Blowdown Tunnel.

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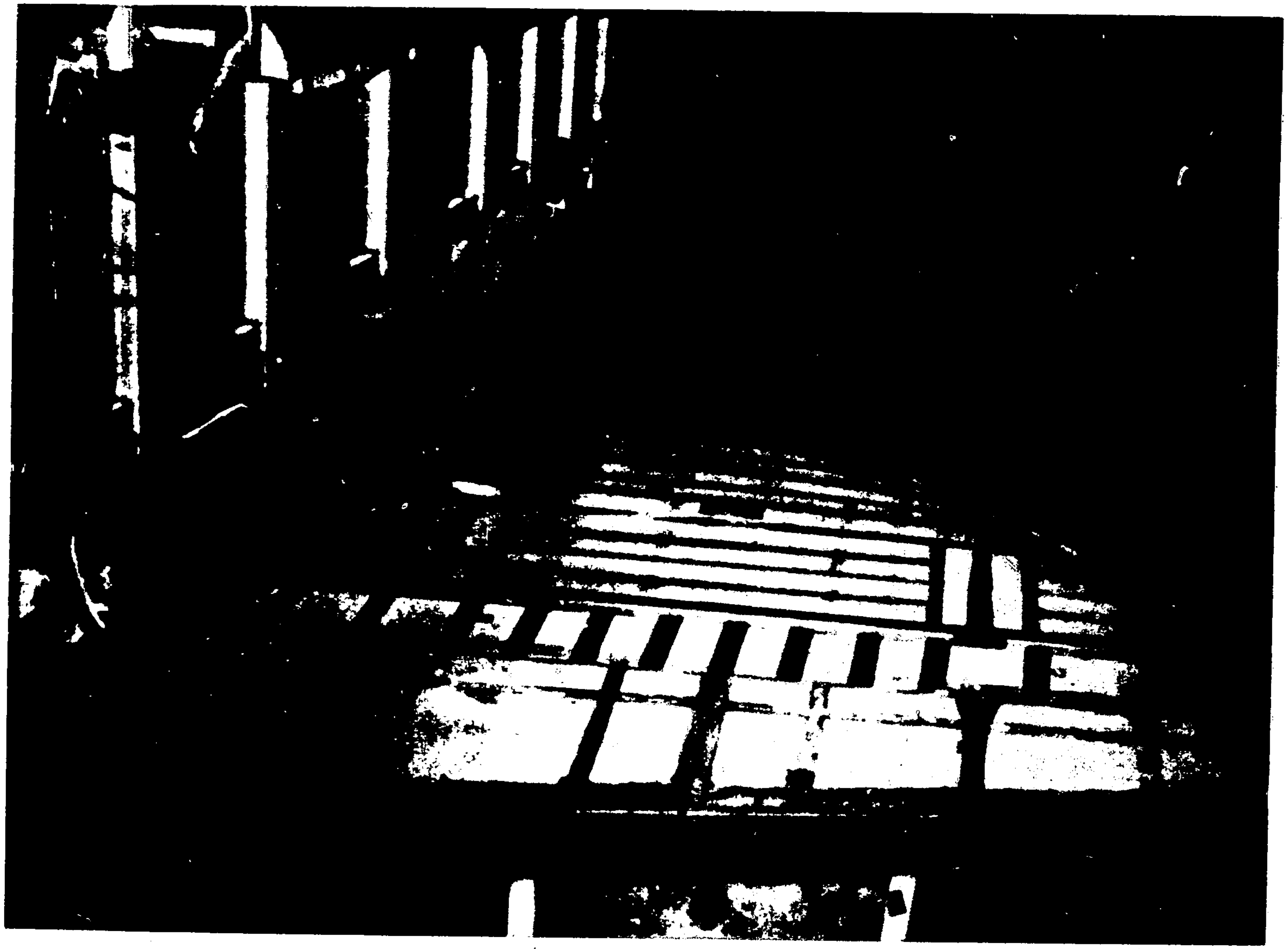
(b) Model assembled in tunnel.

Figure 5.32. - Concluded.

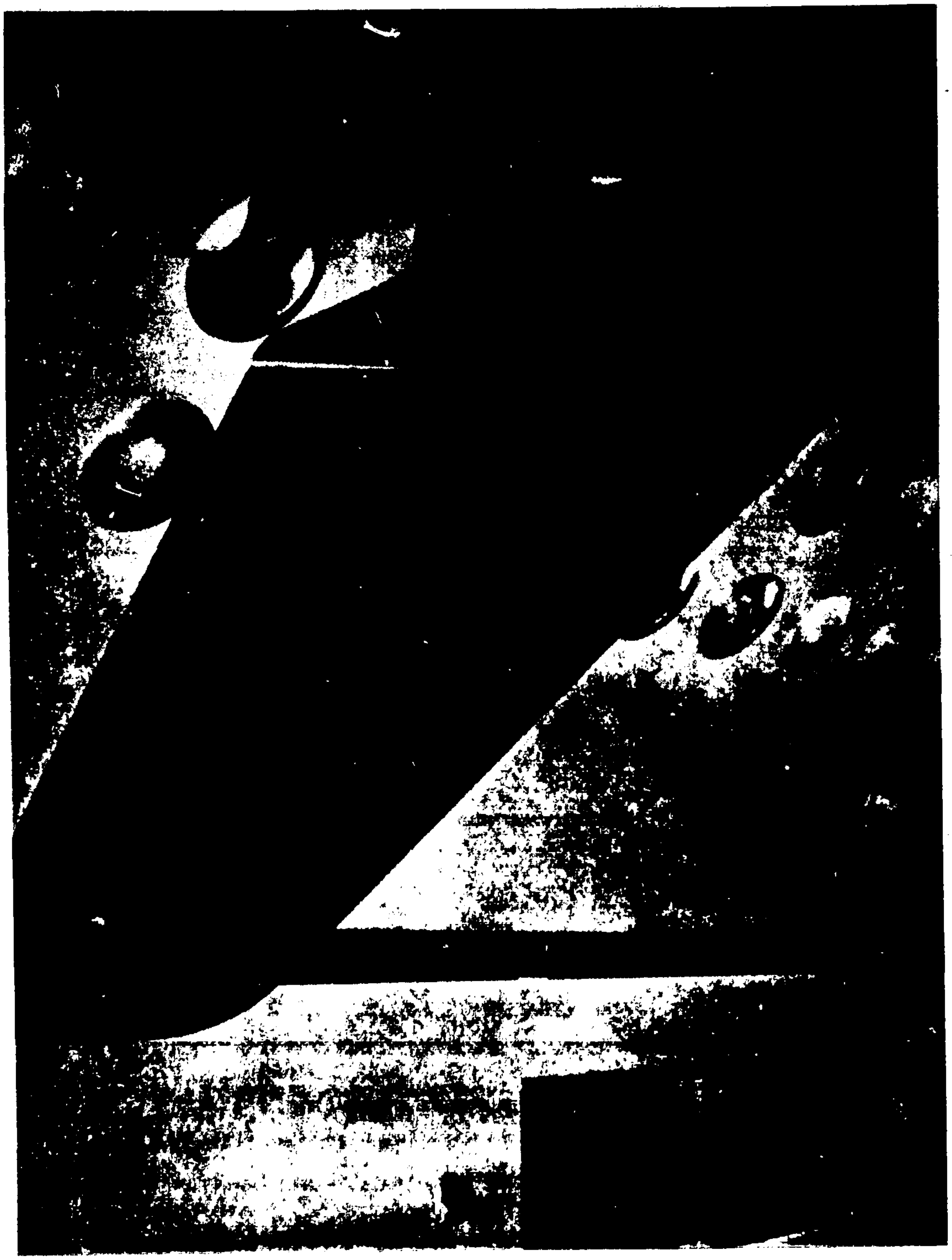


(a) Wing model with skin removed.

Figure 5.33. - Large-scale model tests for flutter boundaries; model no. 59 in the LaRC 16-ft Transonic Dynamics Tunnel.

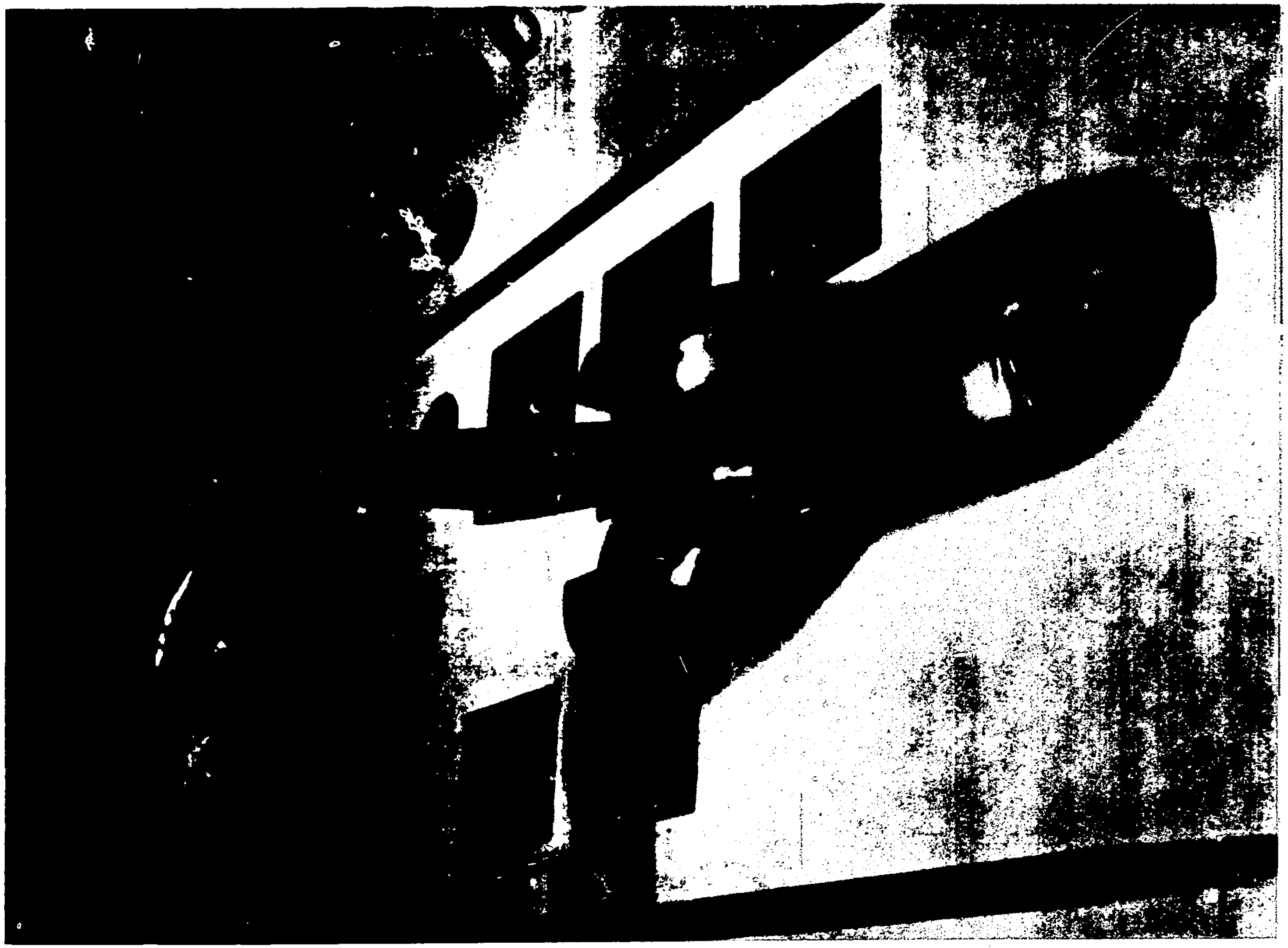
(b) Wing model installed.

Figure 5.33. - Continued.



(c) Vertical tail model with skin removed.

Figure 5.33. - Continued.



(d) Vertical tail model installed.

Figure 5.33. - Concluded.

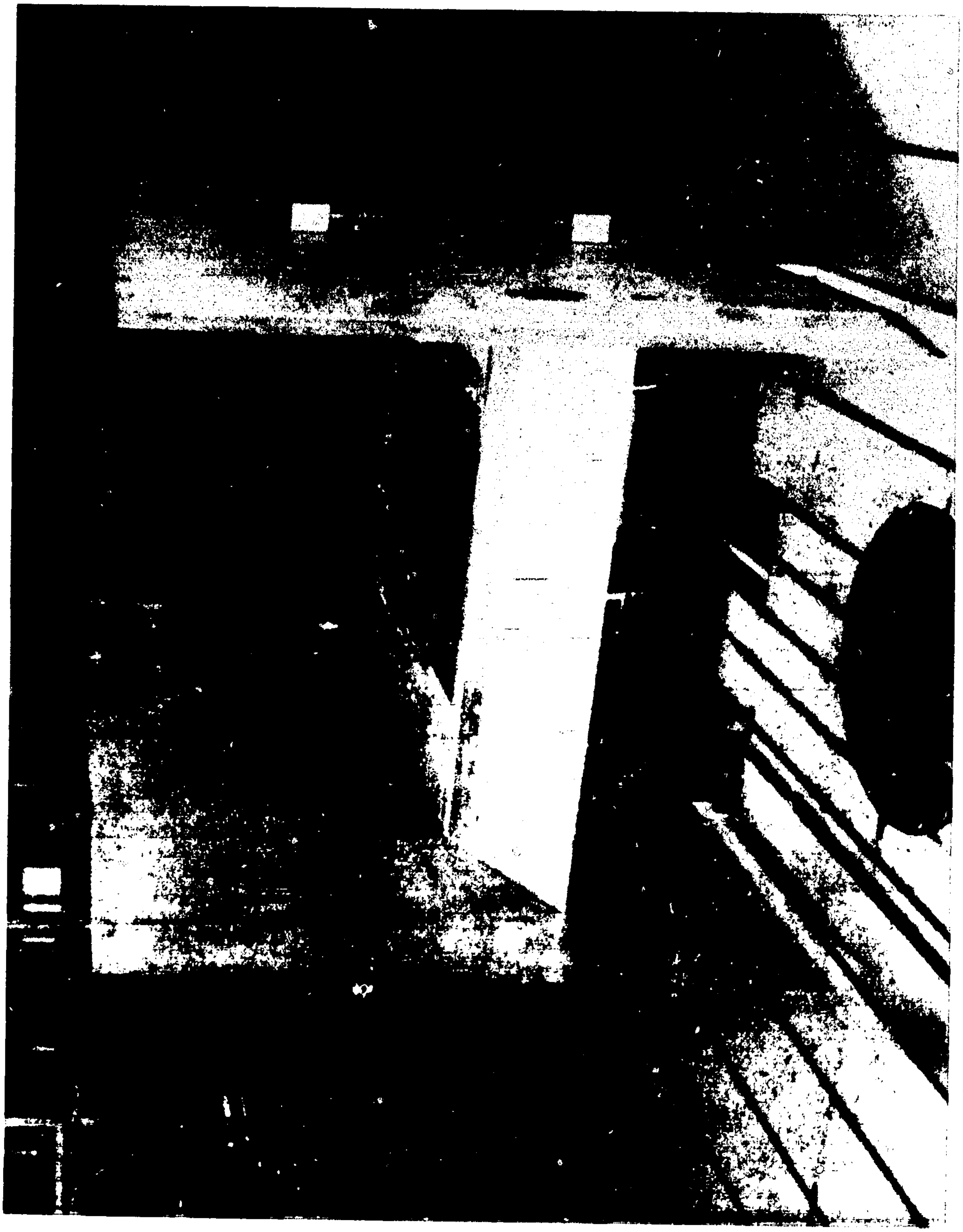
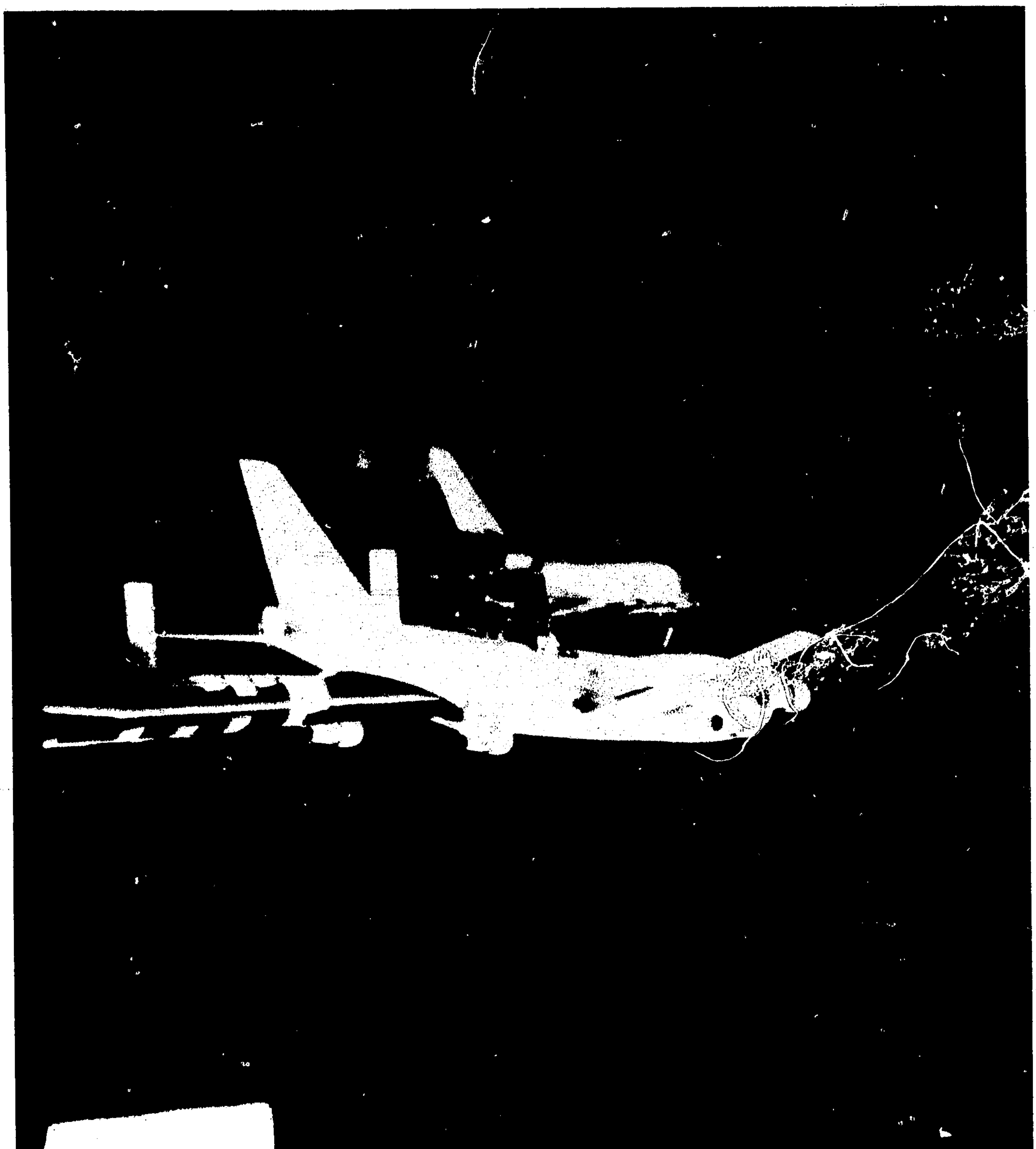


Figure 5.34. - Structural test panel with TPS tiles; model no. 81 in the ARC 11-ft Transonic Tunnel.

(a) Orbiter tailcone configuration.

Figure 5.35. - Mated carrier model no. 48 installed in the ARC 14-ft Transonic Wind Tunnel.



(b) Orbiter tailcone off configuration.

Figure 5.35 - Concluded.

Figure 5.36. - Shuttle Carrier Aircraft separation test; model no. 48 in the ARC 14-ft Transonic Wind Tunnel.

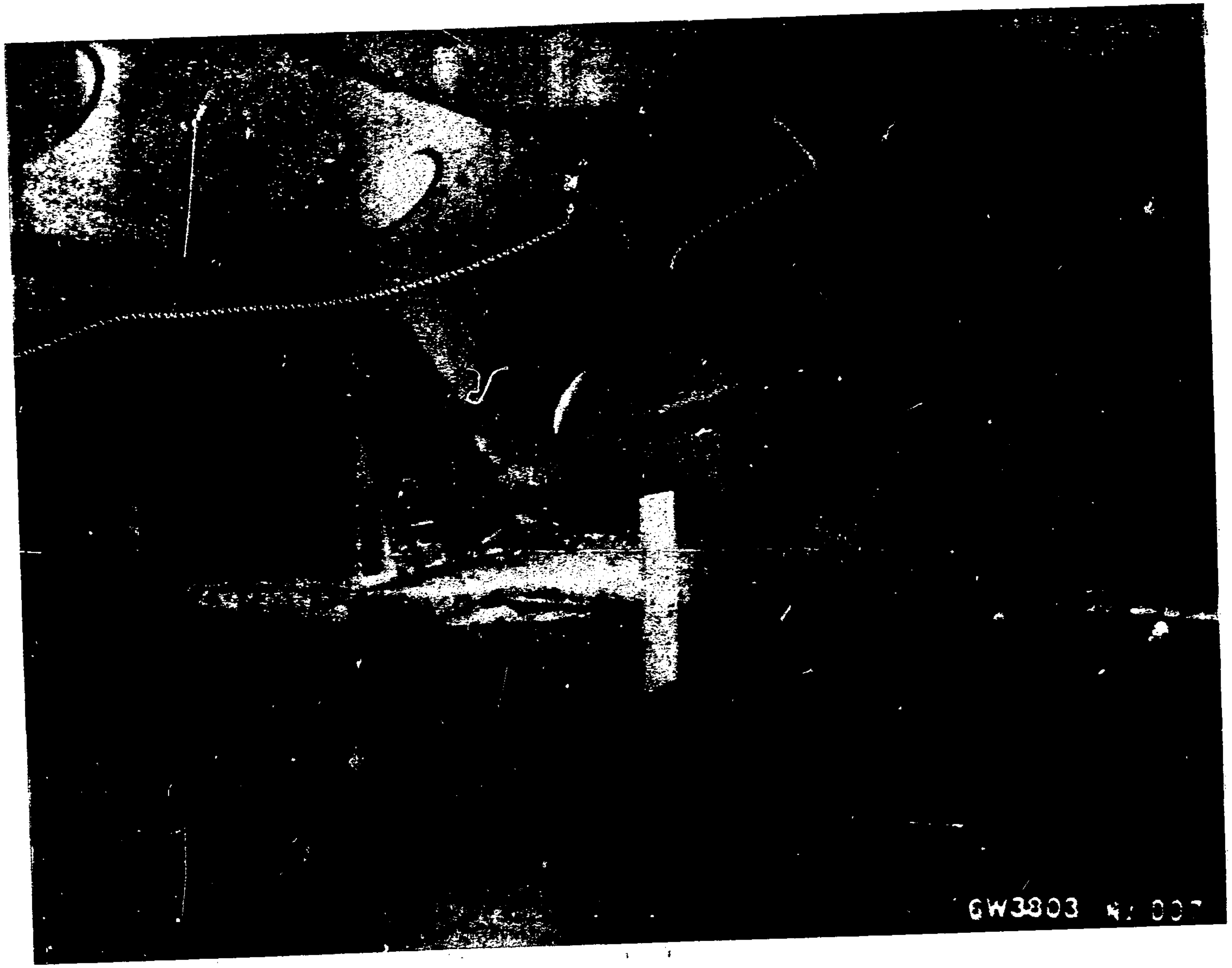


Figure 5.37. - Aeroelastic (tail) buffet test; model no.8 in the University of Washington Low-Speed Wind Tunnel.

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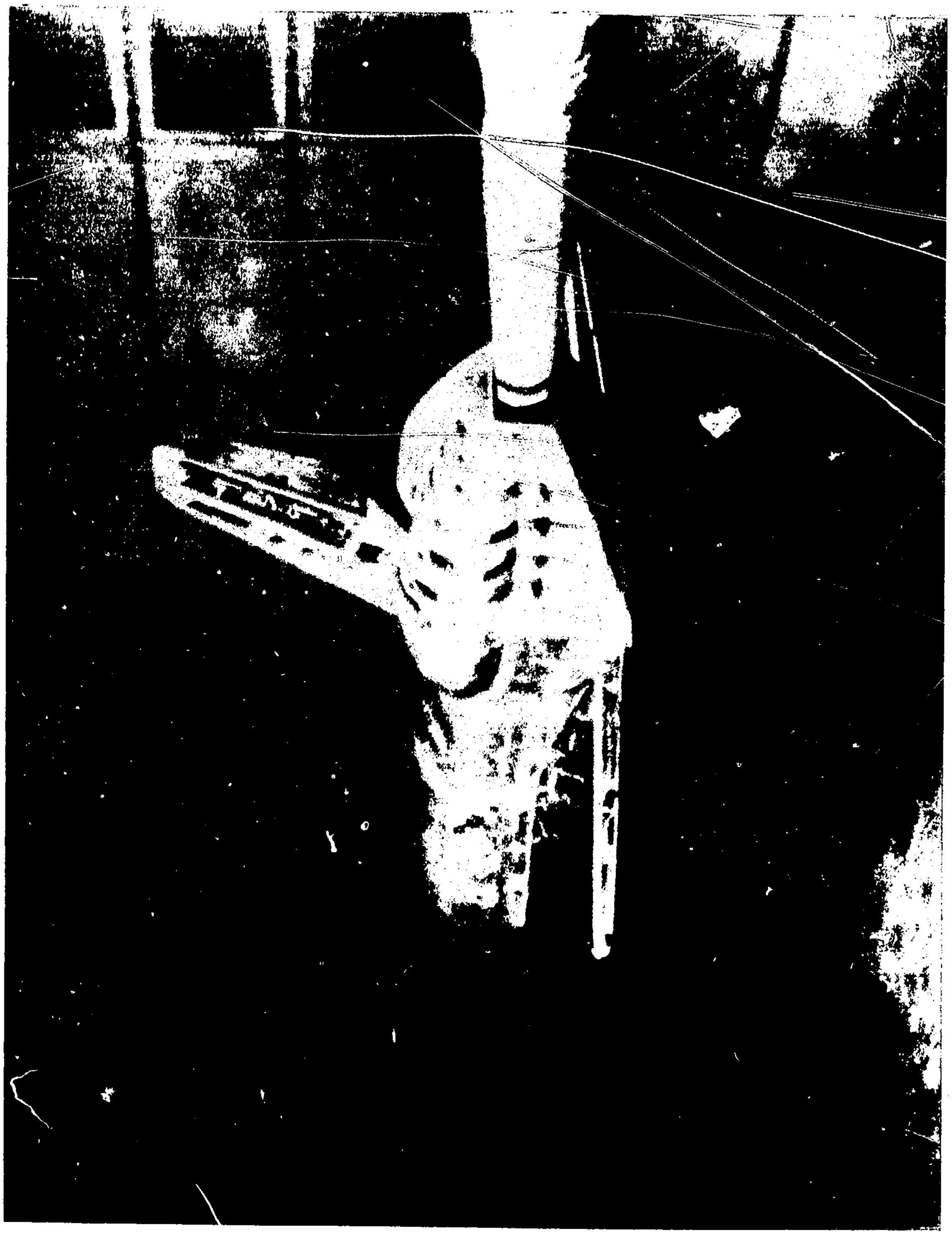

(a) Model no. 47 in the Ames 11-ft Transonic Wind Tunnel; straight sting in Orbiter base.

Figure 5.38. - Tailcone testing for support interference.

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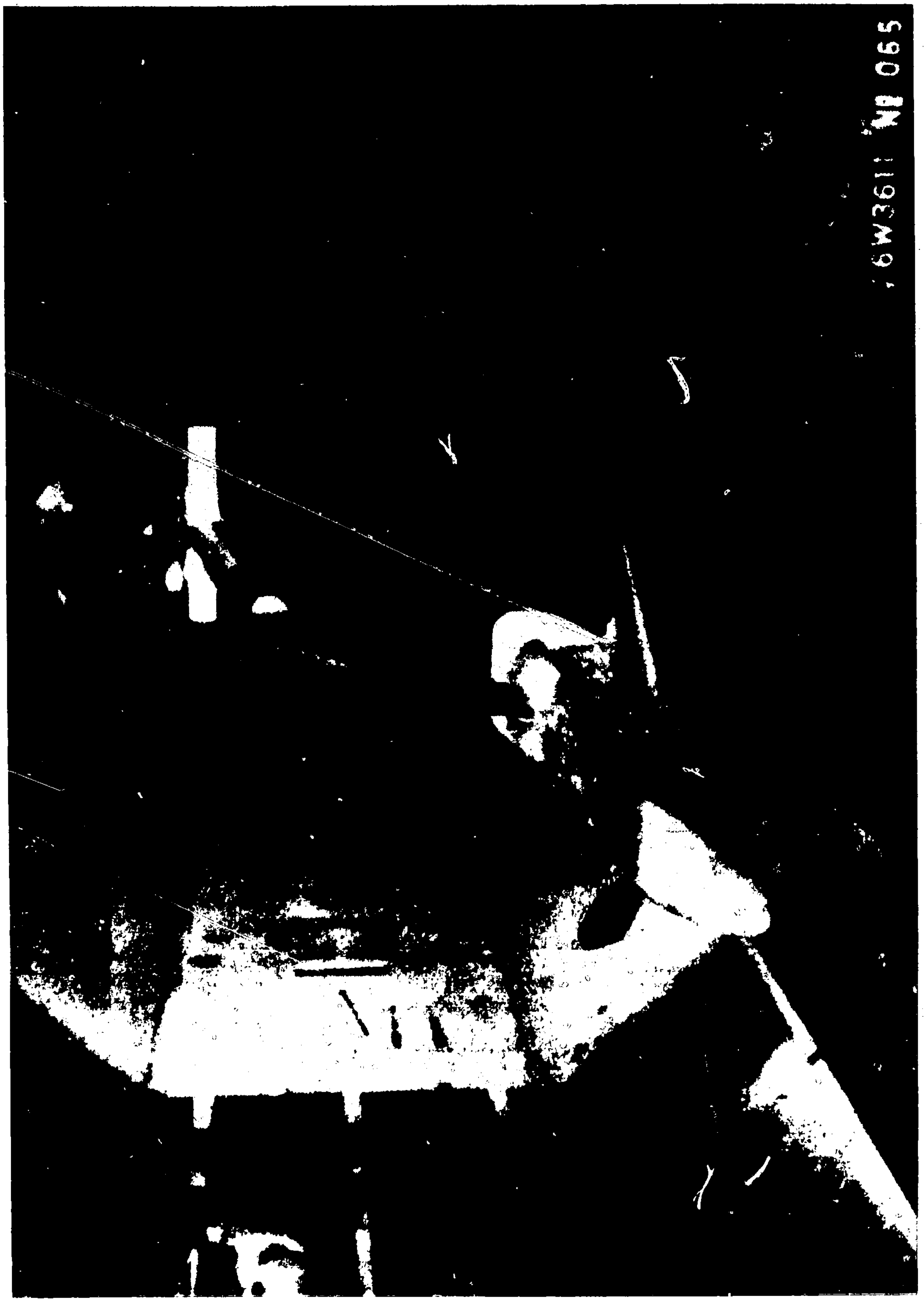


(b) Model no. 47 in the Boeing Transonic Wind Tunnel; vertical tail blade support.

Figure 5.38. - Continued.

(c) Model no. 47 in the Boeing Transonic Wind Tunnel; bottom blade support with dummy vertical tail blade support.

Figure 5.38 - Concluded.

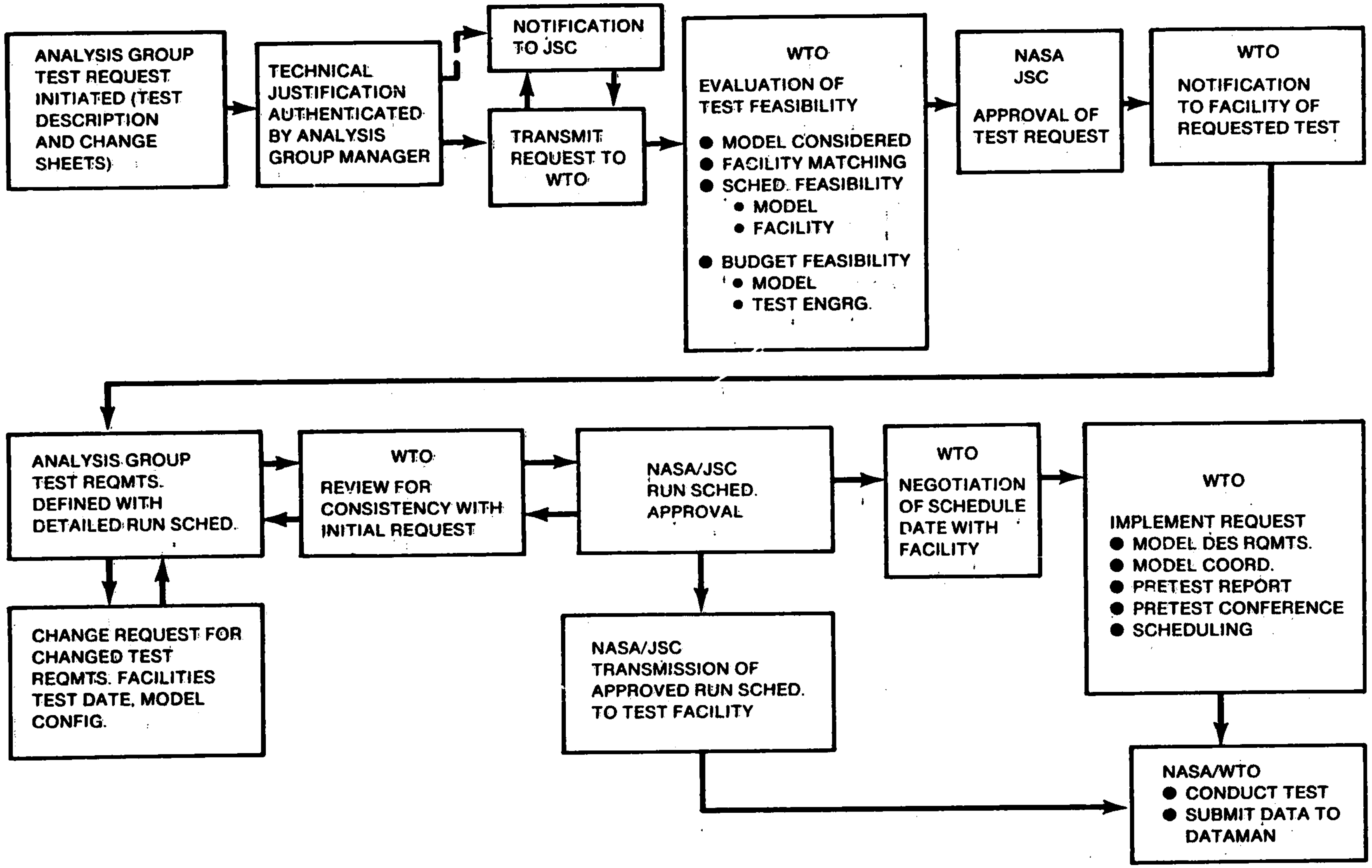


Figure 6.1. - Wind tunnel program management; test approval cycle.

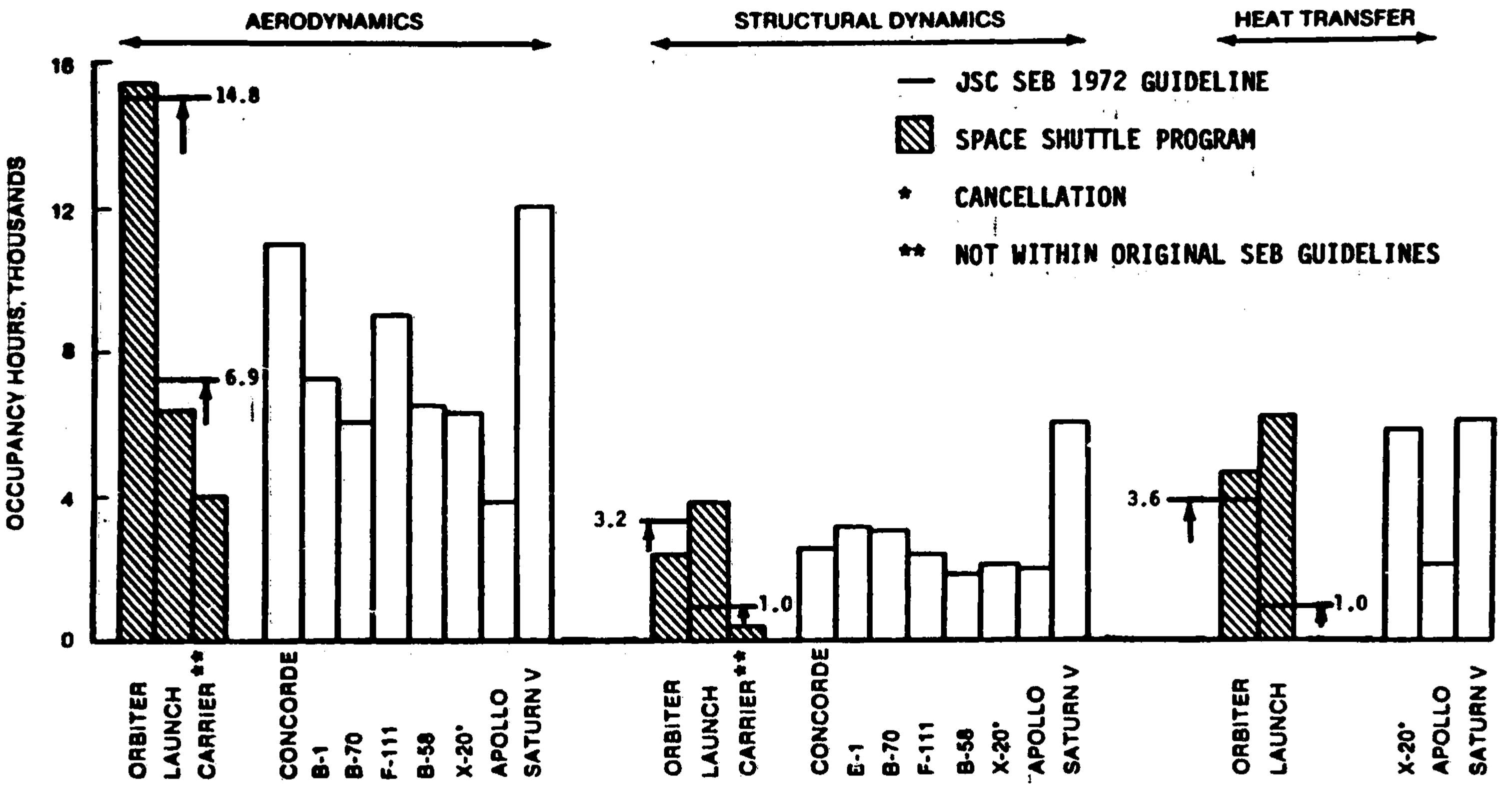
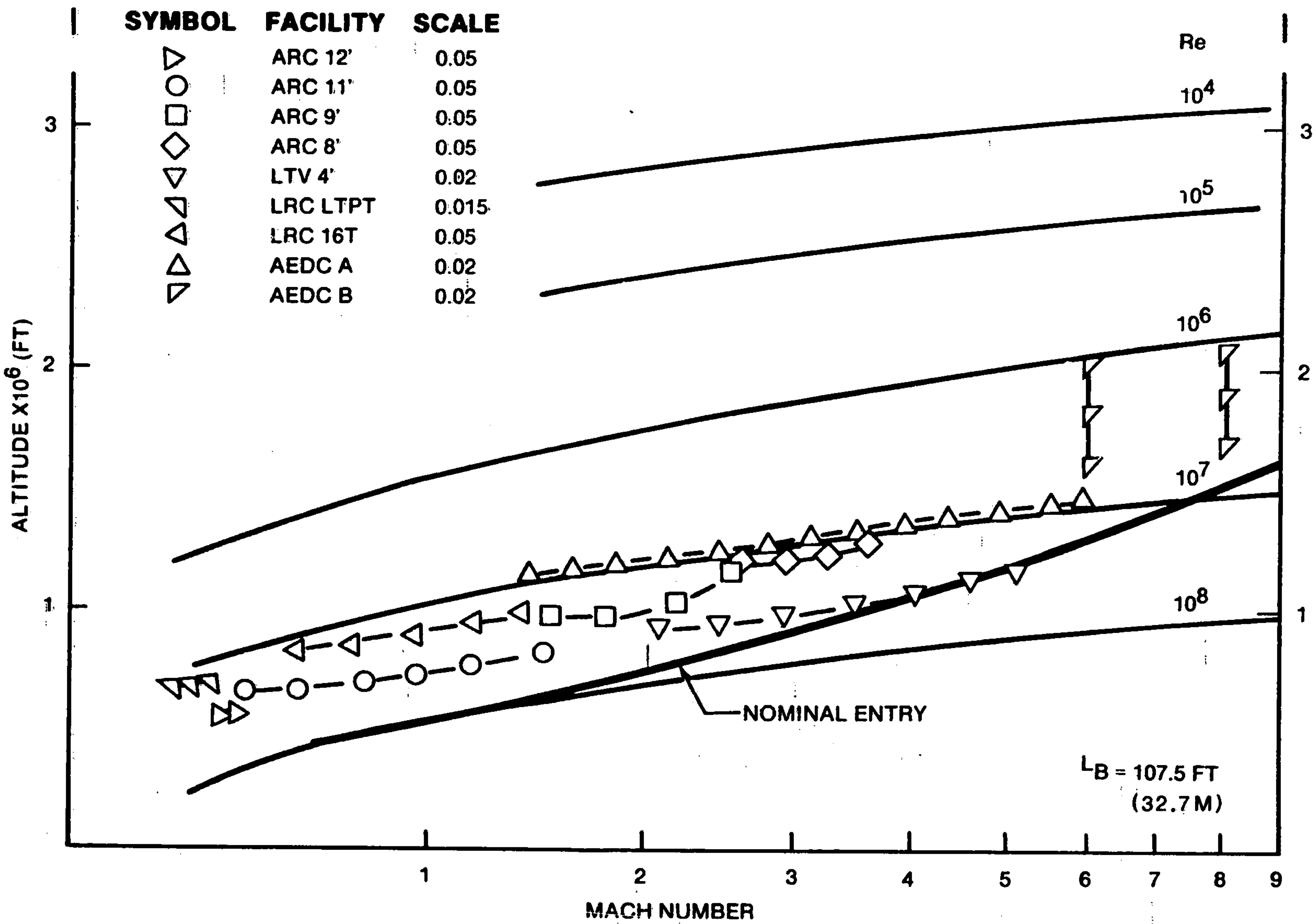


Figure 7.1. - Space Shuttle wind tunnel program comparison.



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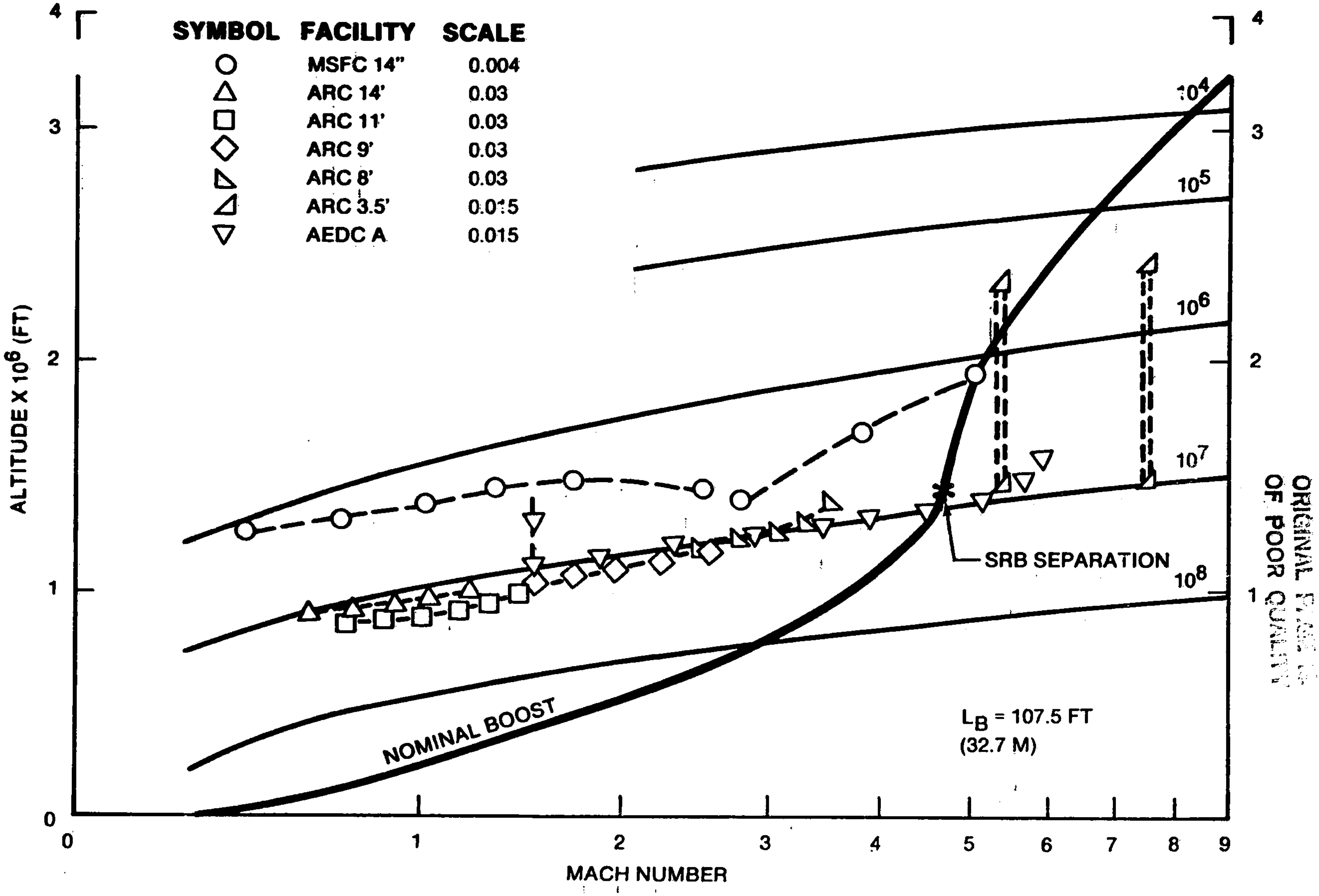
(a) Entry.

Figure 8.1. - Facility Reynolds number simulation capability.



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(b) Ascent.

Figure 8.1. - Concluded.

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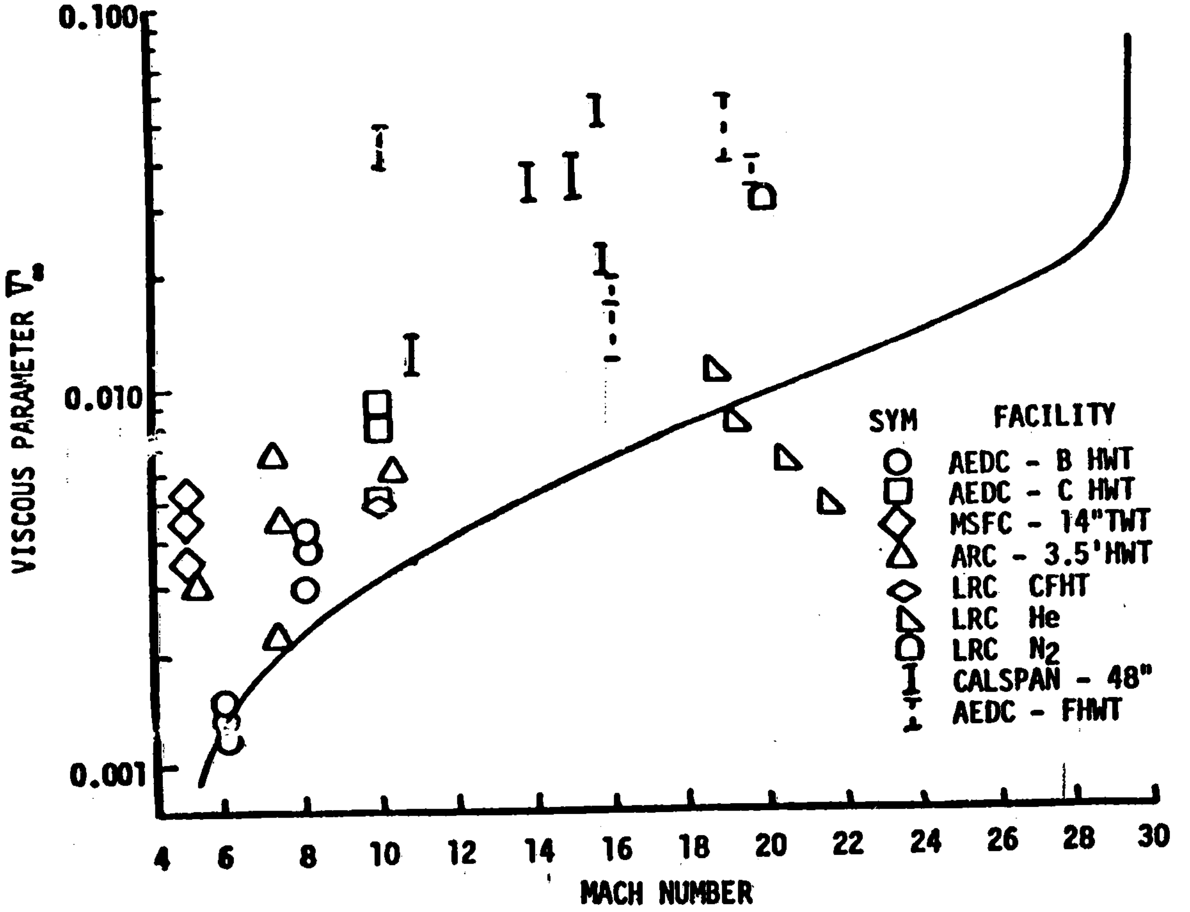
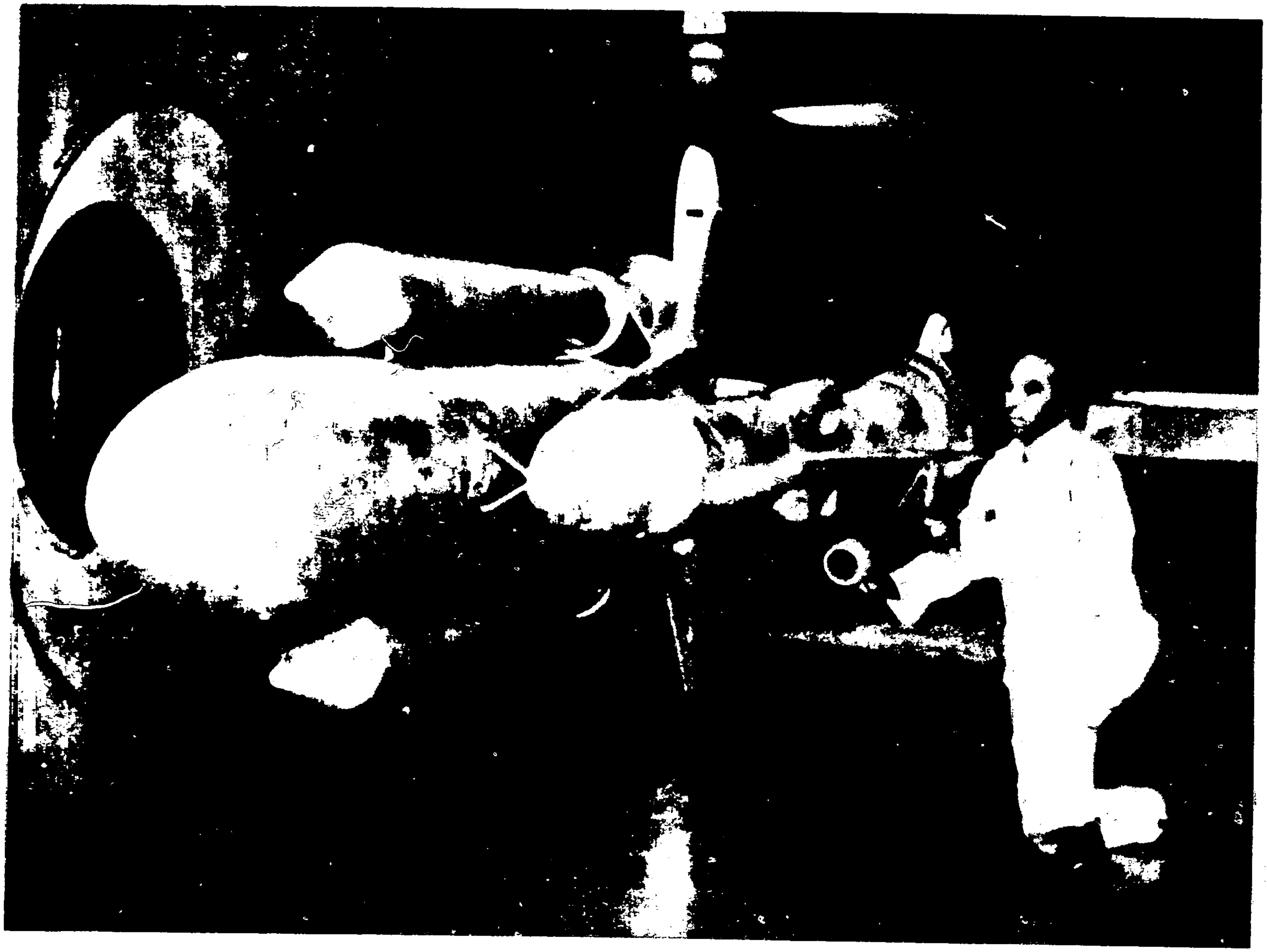


Figure 8.2. - Facility simulation capability in the hypersonic viscous regime.



(a) Model 11 (.040-scale) installed in the ARC 9 by 7-ft Wind Tunnel.

Figure 9.1. - Contrast of model fidelity.

(b) Model 84 (.035-scale) installed in the ARC 9 by 7-ft Wind Tunnel.

Figure 9.1. - Continued.

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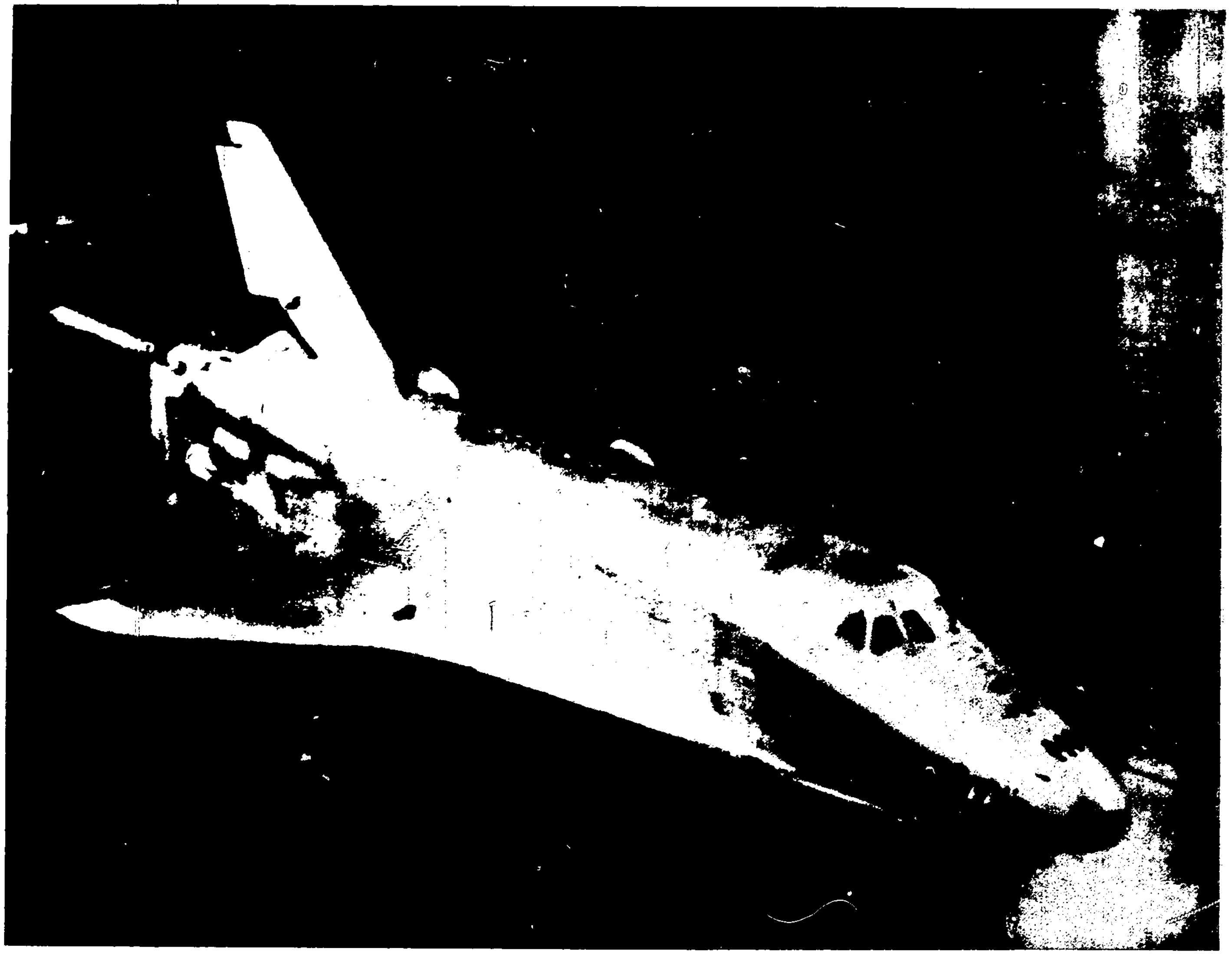
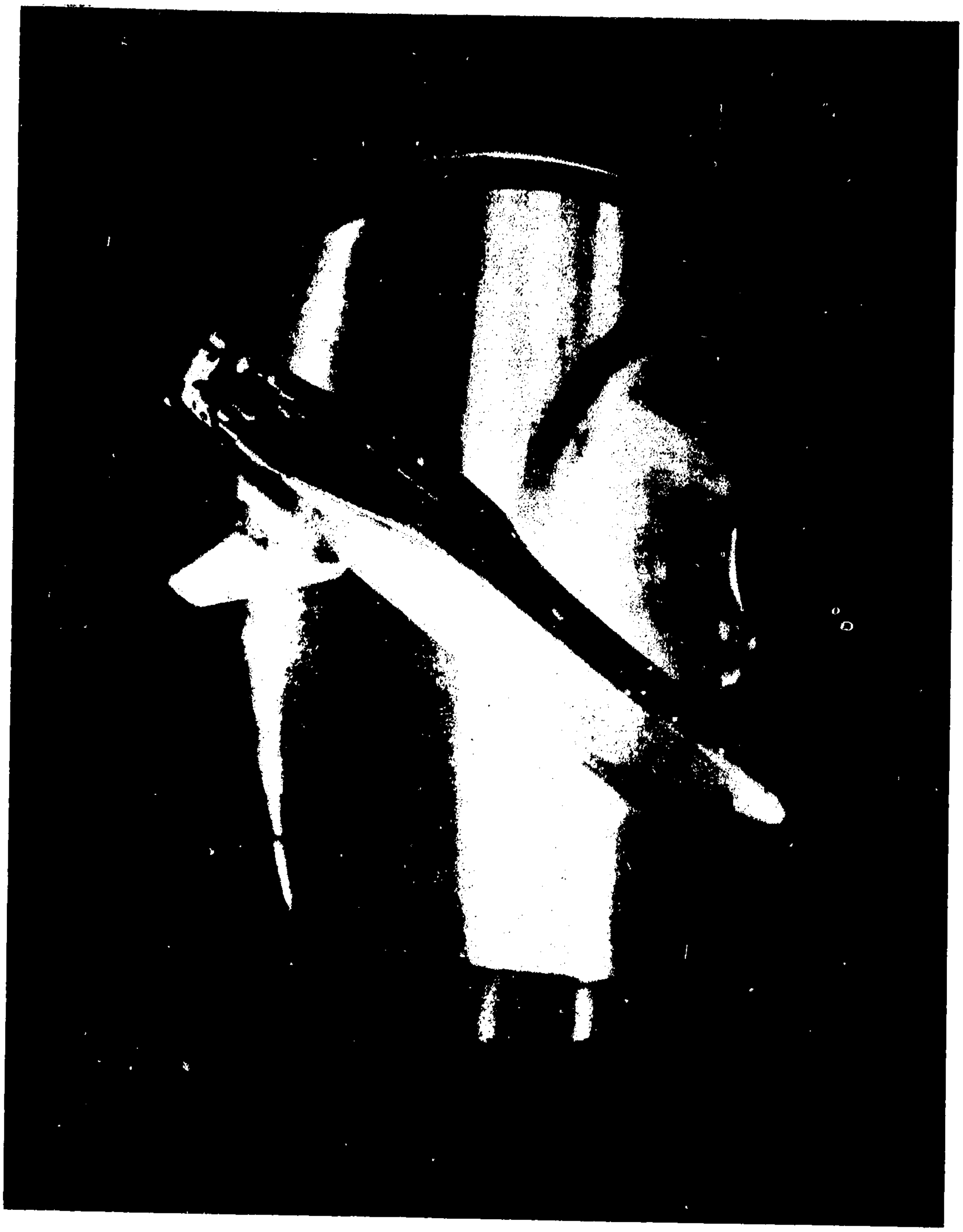


(c) Model 84 (.035-scale) installed in the LeRC 10 by 10-ft Wind Tunnel.

Figure 9.1. - Continued.



(d) Model 84 (.035-scale) installed in the LeRC 10 by 10-ft Wind Tunnel.

Figure 9.1 - Concluded.

APPENDIX A

TABLES OF WIND TUNNEL TESTING BY TEST NUMBER, BY TEST FACILITY, AND BY MODEL NUMBER.

The test number definition, model reference (configuration) definition, and model ID listings are presented in the Chapters VII, III and IX respectively. A summary of the test number definitions is presented here for convenience:

First letter	- O -	Orbiter
	- I -	Integrated Vehicle
	- C -	Carrier Aircraft
	- T -	External Tank
	- S -	Solid Rocket Booster
	- L -	Langley Research Center
	- A -	Ames Research Center
	- M -	Johnson Space Center (formerly the Manned Spacecraft Center)
	- F -	Marshall Space Flight Center
Second Letter	- A -	Aerodynamics Tests
	- H -	Heating Tests
	- S -	Structural Dynamics Tests
Number	-	Chronological Test Order

TABLE A1 - WIND TUNNEL TESTING BY TEST NUMBER - AERODYNAMIC TESTS
TABLE A2 - WIND TUNNEL TESTING BY TEST NUMBER - HEATING TESTS
TABLE A3 - WIND TUNNEL TESTING BY TEST NUMBER - STRUCTURAL DYNAMICS TESTS
TABLE A4 - WIND TUNNEL TESTING BY FACILITY - NASA COMPLEXES
TABLE A5 - WIND TUNNEL TESTING BY FACILITY - OTHER GOVERNMENT COMPLEXES
TABLE A6 - WIND TUNNEL TESTING BY FACILITY - PRIVATE FACILITY COMPLEXES
TABLE A7 - WIND TUNNEL TESTING BY FACILITY - SPACE SHUTTLE PRIME CONTRACTOR COMPLEX
TABLE A8 - WIND TUNNEL TESTING BY FACILITY - UNIVERSITY FACILITIES
TABLE A9 - WIND TUNNEL TESTING BY MODEL

APPENDIX A

TABLE A1 - WIND TUNNEL TESTING BY TEST NUMBER - AERODYNAMIC TESTS

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PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

AUG 01,1984

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
OA1	9*27*72 -	10*07*72	60/ 96	206	ATP	(1)	MSFC 14-IN TRANSONIC	555	2005	PUBLISHED
OA2	9*25*72 -	10*03*72	80/ 69	183	ATP	(2)	RI 7X11-FT LOW SPEED	689	2016	PUBLISHED
OA3	10*24*72 -	11*10*72	200/ 320	214	ATP	(6)	ARC 6X6-FT SUPERSONIC	650	2009	PUBLISHED
OA4	10* 2*72 -	10*17*72	200/ 176	54	ATP	(6)	ARC 3.5-FT HYPERSONIC	147	2007	PUBLISHED
OA5	10*11*72 -	10*19*72	60/ 65	88	ATP-MODIF	(2)	RI 7X11-FT LOW SPEED	690	2017	PUBLISHED
OA6	11*16*72 -	12*06*72	60/ 177	218	PRR	(2)	RI 7X11-FT LOW SPEED	694	2019	PUBLISHED
OA7	11*27*72 -	12*08*72	100/ 100	110	ATP	(6)	LARC UNITARY PLAN	1007	2014	PUBLISHED
OA9	12*18*72 -	01*09*73	60/ 158	192	2A/089B	(2)	RI 7X11-FT LOW SPEED	696	2020	PUBLISHED
OA10	1*30*73 -	02*16*73	120/ 109	300	2A/089B	(2)	RI 7X11-FT LOW SPEED	698	2022	PUBLISHED
OA11A	4* 9*73 -	04*17*73	144/ 176	62	2A/089B	(18)	ARC 3.5-FT HYPERSONIC	157	2044	PUBLISHED
OA11B	5*14*73 -	05*25*73	140/ 160	70	2A/089B	(18)	ARC 3.5-FT HYPERSONIC	160	2059	PUBLISHED
OA12A	4*12*73 -	04*23*73	90/ 103	98	2A/089B	(17)	ARC 11-FT TRANSONIC	707	2032	PUBLISHED
OA12C	5* 2*73 -	05*10*73	60/ 60	46	2A/089B	(17)	ARC 8X7-FT SUPERSONIC	707	2032	PUBLISHED
OA14	2*28*73 -	03*15*73	100/ 151	196	2A/089B	(2)	RI 7X11-FT LOW SPEED	700	2030	PUBLISHED
OA16	3*19*73 -	04*17*73	130/ 320	475	2A/089B	(2)	RI 7X11-FT LOW SPEED	701	2038	PUBLISHED
OA17-1	6*18*73 -	07*06*73	60/ 124	65	3/139B	(42)	LARC LOW TURBULANCE PRESSURE	138	2058	PUBLISHED
OA17-2	6*18*73 -	07*06*73	20/ 100	55	2A/089B	(18)	LARC LOW TURBULANCE PRESSURE	138	2058	PUBLISHED
OA18	5* 8*73 -	05*17*73	100/ 114	189	3/139B	(43)	RI 7X11-FT LOW SPEED	704	2045	PUBLISHED
OA20A	9*10*73 -	09*13*73	50/ 40	29	4/140A,B	(49)	LARC UNITARY PLAN	1057	2083	PUBLISHED
OA20B	4* 8*74 -	04*12*74	50/ 43	30	4/140A,B	(49)	LARC UNITARY PLAN	1097	2163	PUBLISHED
OA20C	11* 5*73 -	11*08*73	40/ 35	19	4/140A,B	(49)	LARC UNITARY PLAN	1057	2147	PUBLISHED
OA21A	5*21*73 -	06*04*73	100/ 72	348	3/139B	(43)	RI 7X11-FT LOW SPEED	705	2053	PUBLISHED
OA21B	6*21*73 -	06*25*73	40/ 55	99	3/139B W/CANS	(43)	RI 7X11-FT LOW SPEED	705	2053	PUBLISHED
OA22A	9*12*73 -	09*14*73	20/ 21	24	4/140A,B	(47)	ARC 11-FT TRANSONIC	716	2130	PUBLISHED
OA22B	9*19*73 -	09*20*73	40/ 31	30	4/140A,B	(47)	ARC 9X7-FT SUPERSONIC	716	2131	PUBLISHED

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TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(IC)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
OA23-1	7*19*73	07*31*73	80/ 54	23	3A/140A	(49)	ARC 3.5-FT HYPERSONIC	168	2071	PUBLISHED
OA23-2	7*26*73	07*31*73	80/ 90	39	3/139B	(32)	ARC 3.5-FT HYPERSONIC	168	2071	PUBLISHED
OA25	9*14*73	09*21*73	80/ 88	156	4/140A,B	(49)	LARC 8-FT TRANSONIC PRESSURE	661	2089	PUBLISHED
OA26	11*17*73	12*04*73	64/ 140	27	4/140A,B	(36)	ARC 3.5-FT HYPERSONIC	180	2124	PUBLISHED
OA36	2*25*74	03*01*74	80/ 80	38	4/140A,B	(49)	ARC 3.5-FT HYPERSONIC	187	2162	PUBLISHED
OA37	1* 7*74	01*25*74	80/ 103	112	4/140A,B	(47)	RI 7X11-FT LOW SPEED	719	2140	PUBLISHED
OA43	4*18*73	05*04*73	128/ 160	137	2A/089B	(18)	ARC 6X6-FT SUPERSONIC	706	2050	PUBLISHED
OA44-1	6* 1*73	06*08*73	40/ 54	47	2A/089B	(18)	LARC UNITARY PLAN	1035	2057	PUBLISHED
OA44-2	6*11*73	06*15*73	40/ 54	36	3/139B	(42)	LARC UNITARY PLAN	1035	2057	PUBLISHED
OA45	2*21*73	02*28*73	80/ 86	171	2A/089B	(2)	RI 7X11-FT LOW SPEED	699	2021	PUBLISHED
OA47	3*28*73	04*05*73	116/ 116	245	2A/089B	(13)	MSFC 14-IN TRANSONIC	568	2029	PUBLISHED
OA48	5*25*73	6*11*73	100/ 166	364	3/139B,W/CANS	(34)	MSFC 14-IN TRANSONIC	574	2055	PUBLISHED
OA49	10*18*73	11*09*73	198/ 170	415	4/140A,B	(34)	MSFC 14-IN TRANSONIC	581	2095	PUBLISHED
OA53A	11*19*73	11*27*73	128/ 128	267	4/140A,B	(47)	ARC 11-FT TRANSONIC	747	2128	PUBLISHED
OA53B	11*12*73	11*16*73	60/ 160	103	4/140A,B	(47)	ARC 9X7-FT SUPERSONIC	747	2178	PUBLISHED
OA53C	11*28*73	12*06*73	60/ 159	159	4/140A,B	(47)	ARC 8X7-FT SUPERSONIC	747	2185	PUBLISHED
OA57A	8* 6*73	8*17*73	100/ 103	61	2A/089B	(2)	RI 7X11-FT LOW SPEED	709	2074	PUBLISHED
OA57B	9*15*73	09*17*73	40/ 123	72	2A/089B	(2)	RI 7X11-FT LOW SPEED	713	2080	PUBLISHED
OA58	6* 4*73	06*18*73	80/ 76	38	3/139B	(42)	ARC 3.5-FT HYPERSONIC	163	2060	PUBLISHED
OA59	3*13*74	3*21*74	120/ 293	150	4/140A,B	(49)	ARC 6X6-FT SUPERSONIC	709	2159	PUBLISHED
OA62A	10* 5*73	10*23*73	120/ 195	98	4/140A,B	(43)	RI 7X11-FT LOW SPEED	715	2097	PUBLISHED
OA62B	11*13*73	12*06*73	100/ 240	448	4/140A,B	(43)	RI 7X11-FT LOW SPEED	717	2104	PUBLISHED
OA63	9*25*73	9*28*73	64/ 80	98	4/140A,B	(36)	ARC 6X6-FT SUPERSONIC	630	2077	PUBLISHED
OA64	10*30*73	10*31*73	50/ 30	28	4/140A,B	(36)	LARC UNITARY PLAN	1063	2108	PUBLISHED
OA66	6*20*73	6*29*73	60/ 60	44	3A/140A	(49)	RI 7-FT TRISONIC	276	2061	PUBLISHED

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OA69	8*28*73 -	09*01*73	80/ 71	205	3/139B	(43)	RI 7X11-FT LOW SPEED	711	2081	PUBLISHED
OA70	7*20*73 -	7*26*73	30/ 40	66	3/139B	(42)	LARC UNITARY PLAN	1043	2073	PUBLISHED
OA71A	7*27*73 -	08*03*73	50/ 62	52	2A/089B	(2)	RI 7X11-FT LOW SPEED	708	2068	PUBLISHED
OA71C	9* 4*73 -	09*14*73	100/ 139	71	3/139B	(43)	RI 7X11-FT LOW SPEED	712	2086	PUBLISHED
OA72	7*30*73 -	08*24*73	40/ 176	42	3A/139B	(34)	LARC 22-IN HELIUM	415	2092	PUBLISHED
OA73	7*11*73 -	7*18*73	60/ 96	37	3/139B	(42)	ARC 3.5-FT HYPERSONIC	167	2082	PUBLISHED
OA77	11*27*73 -	12*01*73	40/ 32	124	4/140A,B	(49)	AEDC B / HYPERSONIC	474	2134	PUBLISHED
OA78	12* 3*73 -	12*04*73	20/ 16	56	4/140A,B	(49)	AEDC C / HYPERSONIC	474	2134	PUBLISHED
OA79	8* 1*74 -	08*03*74	24/ 23	79	4/140A,B(MOD)	(49)	AEDC B / HYPERSONIC	71A	2196	PUBLISHED
OA81	11*28*73 -	12*28*73	104/ 94	48	4/140A,B	(51)	AEDC F / HYPERSONIC	489	2152	PUBLISHED
OA82	8*12*74 -	08*16*74	40/ 48	96	4/140A,B	(32)	LARC 31-IN CONT-FLOW HYP.	113	2195	PUBLISHED
OA83	5* 8*74 -	05*16*74	80/ 160	34	4/140A,B	(36)	ARC 3.5-FT HYPERSONIC	194	2177	PUBLISHED
OA84	12*10*73 -	12*14*73	80/ 115	207	4/140A,B	(49)	LTV 4X4-FT SUPERSONIC	488	2037	PUBLISHED
OA85	10*31*73 -	11*08*73	50/ 60	75	3/139B	(32)	LARC 31-IN CONT-FLOW HYP.	101	2113	PUBLISHED
OA86	10*26*73 -	11*09*73	80/ 174	331	4/140A,B	(43)	RI 7X11-FT LOW SPEED	716	2114	PUBLISHED
OA87	10*15*73 -	10*23*73	80/ 80	30	4/140A,B	(49)	ARC 3.5-FT HYPERSONIC	176	2115	PUBLISHED
OA88	12*11*73 -	12*28*73	60/ 60	191	4/140A,B	(34)	LARC 22-IN HELIUM	7422	2125	PUBLISHED
OA89	7*15*74 -	08*05*74	60/ 143	32	5/140C	(74)	LARC HYPERSONIC NITROGEN	30	2214	PUBLISHED
OA90	3* 4*74 -	03*06*74	25/ 40	43	4/140A,B	(72)	LARC 31-IN CONT-FLOW HYP.	110	2149	PUBLISHED
OA91	10*26*73 -	11*01*73	40/ 40	38	4/140A/B	(49)	RI 7-FT TRISONIC	278	2116	PUBLISHED
OA93	11*18*74 -	11*23*74	80/ 152	15	4/140A,B	(51)	CALSPAN HYPERSONIC SHOCK	737	2238	PUBLISHED
OA98	3*27*74 -	04*03*74	80/ 128	46	4/140A/B	(49)	ARC 3.5-FT HYPERSONIC	190	2167	PUBLISHED
OA99	3*26*74 -	4*12*74	50/ 52	14	3/139B	(21)	LARC 60-FT. VACUUM SPHERE	3289	2172	PUBLISHED
OA100	5*27*75 -	06*14*75	240/ 272	190	OV101(ALT)	(76)	ARC 40X80-FT SUBSONIC	462	2261	PUBLISHED
OA101	9*13*77 -	11*11*77	160/ 160	373	VEH 102	(39)	ARC 12-FT PRESSURE	218	2405	PUBLISHED

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OA102	6*17*74	06*18*74	18/ 18	10	4/140A,B	(36)	LARC 8-FT TRANSONIC PRESSURE	687	2229	PUBLISHED
OA105	2*20*74	2*22*74	16/ 20	50	4/140A,B	(32)	LARC 31-IN CONT-FLOW HYP.	109	2137	PUBLISHED
OA106	12*17*73	12*18*73	20/ 24	18	4/140A,B	(67)	LARC 8-FT TRANSONIC PRESSURE	668	2120	PUBLISHED
OA108	6*24*74	07*09*74	80/ 80	186	5/140C	(74)	MSFC 14-IN TRANSONIC	599	2190	PUBLISHED
OA109	8*26*74	08*29*74	60/ 88	32	5/140C	(74)	LARC 22-IN HELIUM	431	2205	PUBLISHED
OA110	3*15*74	03*20*74	80/ 48	85	4/140A,B	(16)	RI 7X11-FT LOW SPEED	721	2155	PUBLISHED
OA113	8*10*74	10*04*74	24/ 336	108	4/140A,B	(51)	CALSPAN HYPERSONIC SHOCK	184-220	2234	PUBLISHED
OA115A	7*29*74	07*31*74	24/ 28	82	4/140A,B(MOD)	(49)	AEDC A / SUPERSONIC	71A	2198	PUBLISHED
OA116	6*10*74	06*14*74	80/ 80	81	4/140A,B	(49)	LARC 8-FT TRANSONIC PRESSURE	686	2186	PUBLISHED
OA118	4*24*74	04*26*74	48/ 40	54	4/140A,B	(43)	RI 7X11-FT LOW SPEED	724	2139	PUBLISHED
OA119A	6*17*74	06*25*74	20/ 45	45	4/140A,B	(16)	RI 7X11-FT LOW SPEED	726	2187	PUBLISHED
OA119B	8*22*74	09*06*74	60/ 100	213	4/140A,B	(16)	RI 7X11-FT LOW SPEED	730	2203	PUBLISHED
OA123	9* 6*74	09*10*74	40/ 47	141	4/140A,B (ALT)	(43)	RI 7X11-FT LOW SPEED	731	2202	PUBLISHED
OA124	10*14*74	10*23*74	60/ 60	127	4/140A,B	(43)	RI 7X11-FT LOW SPEED	736	2209	PUBLISHED
OA126A	5* 1*78	05*30*78	240/ 131	304	5/140C	(47)	ARC 11-FT TRANSONIC	289	2424	PUBLISHED
OA126B	4*17*78	04*30*78	120/ 97	256	5/140C	(47)	ARC 9X7-FT SUPERSONIC	289	2424	PUBLISHED
OA126C	12* 8*78	12*22*78	80/ 56	134	5/140C	(47)	ARC 8X7-FT SUPERSONIC	289	2424	PUBLISHED
OA129	7* 7*78	07*15*78	40/ 64	477	VEH102	(47)	AEDC 16-FT TRANSONIC	507	2434	PUBLISHED
OA131	9*11*74	09*26*74	80/ 96	109	5/140C	(74)	MSFC 14-IN TRANSONIC	607	2232	PUBLISHED
OA143	11* 6*74	11*11*74	40/ 55	60	4/140A,B	(16)	RI 7X11-FT LOW SPEED	737	2221	PUBLISHED
OA145A	3* 8*77	04*02*77	160/ 480	981	VEH 102	(39)	ARC 11-FT TRANSONIC	118	2380	PUBLISHED
OA145B	4*15*77	05*03*77	80/ 348	240	VEH 102	(39)	ARC 9X7-FT SUPERSONIC	118	2364	PUBLISHED
OA145C	4* 6*77	04*20*77	80/ 100	188	VEH 102	(39)	ARC 8X7-FT SUPERSONIC	118	2389	PUBLISHED
OA146	11*28*78	12*07*78	80/ 116	30	5/140C	(47)	ARC 8X7-FT SUPERSONIC	318	2445	PUBLISHED
OA148	5* 5*75	05*17*75	220/ 264	474	4/140A,B (MOD)	(47)	ARC 11-FT TRANSONIC	073	2254	PUBLISHED

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OA149A	2*24*77	03*04*77	160/ 144	390	5/140C	(47)	ARC 11-FT TRANSONIC	115	2376	PUBLISHED
OA149B	2* 2*77	02*07*77	40/ 168	201	5/140C	(47)	ARC 9X7-FT SUPERSONIC	115	2370	PUBLISHED
OA149C	2*16*77	02*18*77	40/ 144	25	5/140C	(47)	ARC 8X7-FT SUPERSONIC	115	2370	PUBLISHED
OA155	2*10*75	03*07*75	80/ 152	205	4/140A,B (MOD)	(47)	LARC V/STOL	114	2237	IN PROCESS
OA159	6*23*75	07*08*75	160/ 152	50	140A,B/(ALT)	(45)	ARC 12-FT PRESSURE	078	2265	PUBLISHED
OA160	2* 5*75	02*08*75	12/ 12	14	4/140A,B	(51)	AEDC F / HYPERSONIC	28A	2247	PUBLISHED
OA161A	3*10*75	03*20*75	140/ 160	285	140A,B (MOD)	(45)	ARC 11-FT TRANSONIC	094	2245	PUBLISHED
OA161B	3*20*75	03*26*75	24/ 30	49	140A,B (MOD)	(45)	ARC 9X7-FT SUPERSONIC	094	2245	PUBLISHED
OA161C	3*26*75	03*31*75	20/ 22	45	140A,B (MOD)	(45)	ARC 8X7-FT SUPERSONIC	094	2245	PUBLISHED
OA163A	11*24*75	12*09*75	160/ 144	215	4/140A,B	(16)	RI 7X11-FT LOW SPEED	751	2289	PUBLISHED
OA163B	12*21*76	12*23*76	35/ 35	99	4/140A,B	(16)	RI 7X11-FT LOW SPEED	788	2361	PUBLISHED
OA164	11*28*75	12*01*75	80/ 80	22	OV101(ALT)	(76)	ARC 40X80-FT SUBSONIC	473	2499	PUBLISHED
OA169	3*26*76	04*09*76	12/ 43	200	5/140C	(70)	AEDC B / HYPERSONIC	D8A	2320	PUBLISHED
OA171	6* 5*78	06*22*78	180/ 180	35	VEH. 102	(105)	NSWC HYPERSONIC LAB (#9)	1310	2433	PUBLISHED
OA172	12*15*75	01*13*76	120/ 210	122	4/140A,B(ALT)	(43)	RI 7X11-FT LOW SPEED	752	2294	PUBLISHED
OA173	3*15*76	03*26*76	160/ 256	48	140C(ALT)	(45)	ARC 12-FT PRESSURE	180	2304	PUBLISHED
OA174	2* 2*76	02*27*76	240/ 264	165	OV101(ALT)	(76)	ARC 40X80-FT SUBSONIC	479	2302	PUBLISHED
OA175	6*28*76	07*09*76	160/ 240	290	140A,B (ALT)	(47)	ARC 11-FT TRANSONIC	187	2333	PUBLISHED
OA176	3*29*76	04*15*76	60/ 83	113	4/140A,B(ALT)	(43)	RI 7X11-FT LOW SPEED	754	2314	PUBLISHED
OA208	3*30*78	04*06*78	52/ 47	183	VEH. 102	(105)	AEDC B / HYPERSONIC	P5A	2416	PUBLISHED
OA209	3*21*78	03*30*78	65/ 69	324	VEH. 102	(105)	AEDC A / SUPERSONIC	P5A	2415	PUBLISHED
OA220	11*11*75	11*21*75	120/ 110	142	VEH 101 (ADS)	(57)	ARC 14-FT TRANSONIC	150	2286	PUBLISHED
OA221B	11* 8*76	21*15*76	60/ 76	184	ADS PROBES	(99)	ARC 9X7-FT SUPERSONIC	119	2360	PUBLISHED
OA221C	11*15*76	11*22*76	60/ 68	58	ADS PROBES	(99)	ARC 8X7-FT SUPERSONIC	119	2360	PUBLISHED
OA223	11*20*76	11*30*76	40/ 88	13	VEH 102	(39)	RI 7X11-FT LOW SPEED	776	2402	PUBLISHED

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OA224	2•23•76	03•24•76	80/ 304	25	VEH 102 (ADS)	(57)	LARC 16-FT TRANSONIC	312	2329	PUBLISHED
OA228	5•29•76	05•01•76	16/ 23	45	VEH 102 (ADS)	(57)	RI 7X11-FT LOW SPEED	757	2322	PUBLISHED
OA232	2•17•78	03•01•78	80/ 80	281	ADS PROBES	(99)	AEDC 16-FT TRANSONIC	431	2414	PUBLISHED
OA234	6• 7•77	08•11•77	30/ 80	63	ADS PROBES	(99)	LERC 10X10-FT SUPERSONIC	042	2400	PUBLISHED
OA236	5•28•76	06•02•76	10/ 37	204	ADS PROBES	(99)	RI 7X11-FT LOW SPEED	759	2337	PUBLISHED
OA237	1•24•77	01•31•77	60/ 60	32	ADS PROBES	(99)	ARC 40X80-FT SUBSONIC	500	2375	PUBLISHED
OA239	10•25•76	11•09•76	24/ 48	57	ADS PROBES	(99)	RI 7X11-FT LOW SPEED	764	2351	PUBLISHED
OA250	7• 1•77	07•07•77	32/ 34	23	140C(ALT)	(45)	RI 7X11-FT LOW SPEED	775	2392	PUBLISHED
OA251B	4•17•78	04•23•78	40/ 80	90	ADS PROBES	(99)	ARC 9X7-FT SUPERSONIC	282	2421	PUBLISHED
OA251C	5•29•78	06•15•78	40/ 72	96	ADS PROBES	(99)	ARC 8X7-FT SUPERSONIC	282	2421	PUBLISHED
OA252	10•16•79	11•14•79	360/ 360	0	TILE	(107)	ARC 2X2-FT TRANSONIC	382	2473	PUBLISHED
OA253	7• 1•80	07•08•80	80/ 80	139	5/140C	(84)	AEDC 16-FT TRANSONIC	574	2486	PUBLISHED
OA255A	10•13•80	11•07•80	240/ 228	268	OV102	(70)	LARC UNITARY PLAN	1311	2498	PUBLISHED
OA255D	1•12•81	02•02•81	240/ 160	90	OV102	(70)	LARC UNITARY PLAN	1319	2498	PUBLISHED
OA255C	11•24•80	12•15•80	240/ 140	27	OV102	(70)	LARC UNITARY PLAN	1315	2498	PUBLISHED
OA255B	11• 8•80	11•21•80	240/ 132	100	OV102	(70)	LARC UNITARY PLAN	1358	2498	PUBLISHED
OA256	2• 2•81	02•09•81	80/ 32	0	OV102	(70)	LARC 16-FT TRANSONIC	352		UNASSIGNED
OA257	3•12•81	04•20•81	80/ 324	380	VEH 102	(72)	LARC 20-IN HYPERSONIC (M=6)	6559	2465	PUBLISHED
[illegible]	[illegible]	[illegible]	48 [illegible]	[illegible]	[illegible] 102	(106)	AEDC B [illegible] HYPERSONIC	342	2491	PUBLISHED
[illegible]	[illegible]	[illegible]	[illegible]	[illegible]	[illegible]	(72)	AEDC B [illegible] HYPERSONIC	[illegible]	[illegible]	PUBLISHED
[illegible]	[illegible]	[illegible] 78	[illegible]	56	[illegible]	[illegible]	LARC 16-FT TRANSONIC	[illegible]	[illegible]	PUBLISHED
[illegible]	[illegible]	[illegible] 78	[illegible]	[illegible]	[illegible]	[illegible]	LARC 16-FT TRANSONIC	[illegible]	[illegible]	PUBLISHED
[illegible]	[illegible]	04•25•78	[illegible]	[illegible]	[illegible] 102	[illegible]	LARC 16-FT TRANSONIC	325	2419	PUBLISHED
OA400	[illegible]•23•80	05•02•80	120/ 20	[illegible]	3/140C	(47)	ARC 11-FT TRANSONIC	427	2482	PUBLISHED

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PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
IA1A	10*10*72 - 10*19*72	56/ 84	179	ATP	(1)	MSFC 14-IN TRANSONIC	556	2006	PUBLISHED
IA1B	10*19*72 - 11*28*72	150/ 257	361	ATP	(1)	MSFC 14-IN TRANSONIC	545	2010	PUBLISHED
IA2	10*11*72 - 11*03*72	40/ 244	92	PRE-ATP/001	(7)	ARC 9X7-FT SUPERSONIC	616	2013	PUBLISHED
IA3	11* 3*72 - 11*16*72	24/ 41	53	PRE-ATP/001	(10)	RI 7X11-FT LOW SPEED	693	2018	PUBLISHED
IA4	11* 2*72 - 11*17*72	80/ 75	62	PRE-ATP/001	(9)	LTV 4X4-FT SUPERSONIC	458	2015	PUBLISHED
IA6	4*30*73 - 05*03*73	45/ 52	94	2A/089B	(13)	MSFC 14-IN TRANSONIC	571	2039	PUBLISHED
IA7	2*12*73 - 02*23*73	80/ 160	85	PRE-ATP/001	(7)	ARC 11-FT TRANSONIC	686	2024	PUBLISHED
IA8	2*12*73 - 03*12*73	80/ 160	54	ATP	(6)	ARC 14-FT TRANSONIC	711	2173	PUBLISHED
IA9A	4* 2*73 - 04*14*73	90/ 113	118	2A/089B	(17)	ARC 11-FT TRANSONIC	707	2032	PUBLISHED
IA9B	5* 2*73 - 05*09*73	100/ 120	65	2A/089B	(17)	ARC 9X7-FT SUPERSONIC	707	2032	PUBLISHED
IA9C	4*22*73 - 05*01*73	60/ 60	102	2A/089B	(17)	ARC 8X7-FT SUPERSONIC	707	2032	PUBLISHED
IA10	8* 1*73 - 08*03*73	50/ 40	18	3/139B	(32)	ARC 3.5-FT HYPERSONIC	169	2078	PUBLISHED
IA12B	4*23*73 - 05*07*73	120/ 156	63	2A/089B(MOD)	(14)	ARC 9X7-FT SUPERSONIC	710	2048	PUBLISHED
IA12C	7*11*73 - 07*27*73	220/ 220	133	2A/089(MOD)	(14)	ARC 8X7-FT SUPERSONIC	710	2065	PUBLISHED
IA13	7* 5*73 - 07*17*73	40/ 39	762	3/139B	(32)	AEDC A / SUPERSONIC	323	2062	PUBLISHED
IA14A	9* 4*73 - 09*13*73	130/ 151	149	4/140A,B	(47)	ARC 11-FT TRANSONIC	716	2084	PUBLISHED
IA14B	9*14*73 - 09*19*73	48/ 41	66	4/140A,B	(47)	ARC 9X7-FT SUPERSONIC	716	2129	PUBLISHED
IA15	10*10*73 - 10*16*73	64/ 80	25	3/139B	(32)	ARC 3.5-FT HYPERSONIC	175	2102	PUBLISHED
IA16	11*17*73 - 12*04*73	80/ 52	9	4/140A,B	(36)	ARC 3.5-FT HYPERSONIC	180	2124	PUBLISHED
IA17A	3* 6*74 - 03*15*74	40/ 45	997	3/139B	(52)	AEDC B / HYPERSONIC	422	2156	PUBLISHED
IA17B	3*18*74 - 03*19*74	8/ 8	13	3/139B	(52)	AEDC B / HYPERSONIC	422	2230	PUBLISHED
IA18	4* 9*74 - 04*12*74	60/ 64	26	3/139B	(52)	ARC 3.5-FT HYPERSONIC	191	2160	PUBLISHED
IA19A	9*16*74 - 09*23*74	156/ 136	201	5/140C	(88)	ARC 11-FT TRANSONIC	014	2170	PUBLISHED
IA22	5* 3*76 - 05*08*76	52/ 49	750	5/140C	(70)	AEDC B / HYPERSONIC	59A	2327	PUBLISHED
IA29	9*12*73 - 09*25*73	80/ 184	111	4/140A,B	(36)	ARC 6X6-FT SUPERSONIC	630	2077	PUBLISHED

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TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
IA31FA	4* 9*73 - 04*13*73	60/ 60	104	2A/089B	(13)	MSFC 14-IN TRANSONIC	566	2026	PUBLISHED
IA31FB	4*13*73 - 04*30*73	50/ 271	220	2A/089B	(13)	MSFC 14-IN TRANSONIC	570	2028	PUBLISHED
IA31FC	6*21*73 - 07*09*73	32/ 51	145	2A/089B	(13)	MSFC 14-IN TRANSONIC	573	2072	PUBLISHED
IA32F	5* 9*73 - 05*24*73	100/ 180	190	2A/089B	(13)	MSFC 14-IN TRANSONIC	567	2027	PUBLISHED
IA33	5* 9*74 - 07*21*74	256/ 264	270	5/140C	(74)	MSFC 14-IN TRANSONIC	594	2174	PUBLISHED
IA35	11* 1*73 - 11*02*73	60/ 30	22	4/140A,B	(36)	LARC UNITARY PLAN	1063	2108	PUBLISHED
IA36	6*15*73 - 06*22*73	60/ 80	120	2A/089(MOD)	(14)	CALSPAN 8-FT TRANSONIC	053	2064	PUBLISHED
IA37A	7*10*73 - 07*13*73	60/ 36	64	3A/139B	(34)	MSFC 14-IN TRANSONIC	579	2063	PUBLISHED
IA37B	10*15*73 - 10*16*73	16/ 22	42	3A/139B	(34)	MSFC 14-IN TRANSONIC	585	2093	PUBLISHED
IA40	6*23*76 - 06*29*76	26/ 41	346	5/140C	(75)	AEDC A / SUPERSONIC	425	2293	PUBLISHED
IA41	12*11*73 - 12*14*73	80/ 64	86	4/140A,B	(67)	LARC 8-FT TRANSONIC PRESSURE	667	2118	PUBLISHED
IA42A	11*27*73 - 12*04*73	40/ 70	62	4/140A,B	(67)	LARC UNITARY PLAN	1056	2119	PUBLISHED
IA42B	12*17*73 - 12*21*73	60/ 50	42	4/140A,B	(67)	LARC UNITARY PLAN	1073	2119	PUBLISHED
IA43	8*26*74 - 09*03*74	80/ 80	105	4/140A,B	(72)	LARC 8-FT TRANSONIC PRESSURE	693	2204	PUBLISHED
IA44A	8*12*74 - 08*16*74	40/ 50	27	4/140A,B	(72)	LARC UNITARY PLAN	1088	2206	PUBLISHED
IA44B	8*19*74 - 08*23*74	40/ 80	47	4/140A,B	(72)	LARC UNITARY PLAN	1119	2206	PUBLISHED
IA48	7*18*73 - 07*21*73	20/ 24	40	3A/139B	(34)	MSFC 14-IN TRANSONIC	580	2063	PUBLISHED
IA52	10*11*73 - 10*17*73	16/ 28	27	3A/139B	(34)	MSFC 14-IN TRANSONIC	584	2042	PUBLISHED
IA53	12*20*73 - 01*04*74	40/ 36	45	2A/089B	(13)	MSFC 14-IN TRANSONIC	588	2123	PUBLISHED
IA57	11*20*73 - 11*20*73	10/ 9	10	3/139,089B	(32)	AEDC A / SUPERSONIC	422	2112	PUBLISHED
IA58	2*11*74 - 02*13*74	32/ 40	34	3/139,089B	(32)	LARC 31-IN CONT-FLOW HYP.	107	2133	PUBLISHED
IA60	2*14*74 - 02*20*74	15/ 36	55	3/139,089B	(32)	LARC 31-IN CONT-FLOW HYP.	108	2137	PUBLISHED
IA61A	1*30*[illegible] - 01*31*74	10/ 10	88	3/139,089B	(32)	AEDC A / SUPERSONIC	422	2143	PUBLISHED
IA61B	2*26*74 - 02*26*74	8/ 8	9	3/139,089B	(52)	AEDC A / SUPERSONIC	21AA	2226	PUBLISHED
IA62F	11*15*73 - 11*19*73	16/ 19	33	4/140A,B	(34)	MSFC 14-IN TRANSONIC	589	2103	PUBLISHED

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TEST NO.	TESTING SCHED.	COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
IA68	1*18*74 -	01*29*74	32/ 36	34	2A/089B	(13)	RI 7-FT TRISONIC	281	2144	PUBLISHED
IA69	1*10*74 -	01*14*75	24/ 25	14	4/140A,B	(67)	RI 7-FT TRISONIC	280	2122	PUBLISHED
IA70	5* 3*74 -	05*24*74	80/ 161	173	4/140A,B	(49)	RI 7-FT TRISONIC	282	2175	PUBLISHED
IA71A-1	12*11*74 -	12*17*74	20/ 40	40	5/140C	(77)	MSFC 14-IN TRANSONIC	610	2227	PUBLISHED
IA71A-2	12*11*74 -	12*17*74	20/ 17	29	5/140C	(74)	MSFC 14-IN TRANSONIC	610	2227	PUBLISHED
IA71B-1	12*19*74 -	01*09*75	40/ 64	90	5/140C	(77)	MSFC 14-IN TRANSONIC	610	2227	PUBLISHED
IA71B-2	12*19*74 -	01*09*75	16/ 16	41	5/140C	(74)	MSFC 14-IN TRANSONIC	610	2227	PUBLISHED
IA72	5*19*75 -	05*31*75	120/ 200	176	5/140C	(88)	ARC 11-FT TRANSONIC	072	2258	PUBLISHED
IA80	11* 4*74 -	11*08*74	100/ 144	380	5/140C	(88)	ARC 11-FT TRANSONIC	023	2212	PUBLISHED
IA81A	7*26*74 -	08*27*74	84/ 184	99	4/140A,B (MOD)	(47)	ARC 11-FT TRANSONIC	019	2169	PUBLISHED
IA81B	8* 9*74 -	08*22*74	60/ 208	88	4/140A,B (MOD)	(47)	ARC 9X7-FT SUPERSONIC	019	2194	PUBLISHED
IA82B	1*28*75 -	02*04*75	70/ 132	286	5/140C	(75)	ARC 9X7-FT SUPERSONIC	044	2231	PUBLISHED
IA82C	11*11*74 -	11*15*74	80/ 92	240	5/140C	(75)	ARC 8X7-FT SUPERSONIC	044	2219	PUBLISHED
IA87	7*18*74 -	07*20*74	24/ 23	90	3/139B	(52)	AEDC A / SUPERSONIC	60A	2192	PUBLISHED
IA93	5*10*76 -	05*14*76	80/ 96	255	5/140C	(72)	LARC 8-FT TRANSONIC PRESSURE	749	2326	PUBLISHED
IA94A	4*18*76 -	04*23*76	40/ 60	92	5/140C	(72)	LARC UNITARY PLAN	1152	2323	PUBLISHED
IA94B	4*26*76 -	05*04*76	80/ 84	144	5/140C	(72)	LARC UNITARY PLAN	1177	2324	PUBLISHED
IA105A	9* 2*77 -	11*20*77	290/ 281	885	5/140C	(47)	AEDC 16-FT TRANSONIC	470	2398	PUBLISHED
IA105B	1* 9*78 -	02*01*78	100/ 258	143	5/140C	(47)	ARC 9X7-FT SUPERSONIC	242	2413	PUBLISHED
IA109	7*26*70 -	8* 8*74	40/ 100	19	2A/089B	(25)	MSFC IMPULSE BASE FLOW FAC.	27	2382	PUBLISHED
IA110-1	7* 8*74 -	07*11*74	50/ 60	79	4/140A,B	(49)	ARC 9X7-FT SUPERSONIC	052	2189	PUBLISHED
IA110-2	7* 8*74 -	07*11*74	30/ 20	17	4/140A,B	(67)	ARC 9X7-FT SUPERSONIC	052	2189	PUBLISHED
IA111	3*21*75 -	03*28*75	36/ 33	1475	3/139B	(52)	AEDC A / SUPERSONIC	A3A	2242	PUBLISHED
IA114	8*18*75 -	08*22*75	42/ 56	100	5/140C	(52)	AEDC B / HYPERSONIC	C4A	2272	PUBLISHED
IA119	10* 7*77 -	10*31*77	170/ 285	620	5/140C	(88)	ARC 11-FT TRANSONIC	275	2404	PUBLISHED

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IA125-1	4*25*75 - 05*22*75	60/ 93	137	5/140C	(74)	MSFC 14-IN TRANSONIC	622	2253	PUBLISHED
IA125-2	4*25*75 - 05*22*75	40/ 30	50	5/140C(74TS)	(77)	MSFC 14-IN TRANSONIC	622	2253	PUBLISHED
IA131B	11* 3*78 - 11*09*78	48/ 40	0	ET FORETANK	(68)	ARC 9X7-FT SUPERSONIC	283	2462	PUBLISHED
IA131C	3* 5*79 - 03*11*79	48/ 40	0	ET FORETANK	(68)	ARC 8X7-FT SUPERSONIC	283	2462	PUBLISHED
IA132	11*27*78 - 12*14*78	96/ 96	0	ET FORETANK	(68)	AEDC 16-FT TRANSONIC	505	2449	PUBLISHED
IA135A	3* 2*76 - 03*23*76	120/ 146	132	4/140A,B (MOD)	(47)	ARC 11-FT TRANSONIC	144	2306	PUBLISHED
IA135B	3* 5*76 - 03*23*76	60/ 100	50	4/140A,B (MOD)	(47)	ARC 9X7-FT SUPERSONIC	144	2306	PUBLISHED
IA135C	3*12*76 - 03*23*76	20/ 40	5	4/140A,B (MOD)	(47)	ARC 8X7-FT SUPERSONIC	144	2306	PUBLISHED
IA137	4*26*76 - 05*03*76	40/ 56	43	ET FORETANK	(68)	ARC 14-FT TRANSONIC	143	2316	PUBLISHED
IA138	8*21*78 - 09*01*78	70/ 112	224	5/140C	(75)	ARC 9X7-FT SUPERSONIC	246	2438	PUBLISHED
IA140A	6* 1*76 - 08*03*76	64/ 222	23C	5/140C	(74)	MSFC 14-IN TRANSONIC	641	2335	PUBLISHED
IA140B	10* 1*76 - 01*28*77	80/ 279	44	5/140C	(74)	MSFC 14-IN TRANSONIC	646	2335	PUBLISHED
IA141	3*31*76 - 04*05*76	30/ 30	37	5/140C	(72)	RI 7-FT TRISONIC	297	2315	PUBLISHED
IA142	8*11*76 - 08*18*76	78/ 64	1900	5/140C	(75)	AEDC A / SUPERSONIC	K1A	2346	PUBLISHED
IA143	11* 8*76 - 11*13*76	65/ 58	2442	5/140C	(75)	AEDC A / SUPERSONIC	P8A	2354	PUBLISHED
IA144	4* 6*77 - 04*15*77	160/ 200	514	5/140C	(72)	ARC 11-FT TRANSONIC	228	2377	PUBLISHED
IA148	4*27*77 - 05*03*77	52/ 52	272	5/140C	(70)	AEDC B / HYPERSONIC	TOA	2384	PUBLISHED
IA156A	10*28*77 - 11*10*77	96/ 124	575	VEH 102	(89)	AEDC 16-FT TRANSONIC	470	2403	PUBLISHED
IA156B	12*16*77 - 01*06*78	100/ 191	177	VEH 102	(89)	ARC 9X7-FT SUPERSONIC	272	2408	PUBLISHED
IA180	3*26*79 - 03*30*79	48/ 53	37	ET FORETANK	(68)	LARC UNITARY PLAN	1267	2457	PUBLISHED
IA181	12*15*77 - 02*03*78	120/ 120	111	5/140C	(74)	MSFC 14-IN TRANSONIC	649	2406	PUBLISHED
IA182	9*19*78 - 09*20*78	12/ 24	87	5/140C	(47)	AEDC 16-FT TRANSONIC	517	2439	PUBLISHED
IA183	11*15*78 - 11*16*78	12/ 12	40	VEH 102	(89)	AEDC 16-FT TRANSONIC	519	2444	PUBLISHED
IA184	4* 2*79 - 04*13*79	24/ 40	115	5/140C	(47)	ARC 9X7-FT SUPERSONIC	347	2456	PUBLISHED
IA190A	2* 7*80 - 02*21*80	160/ 168	166	5/140C	(47)	ARC 11-FT TRANSONIC	411	2476	IN PROCESS

TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
IA190B	5*20*80 - 02*21*80	120/ 104	294	5/140C	(47)	ARC 9X7-FT SUPERSONIC	411	2476	IN PROCESS
IA191	6*20*80 - 06*27*80	40/ 40	0	FUEL LINE	(112)	ARC 11-FT TRANSONIC	412	2378	PUBLISHED
IA193	2*26*82 - 04*31*82	72/ 720	0	VEH 102	(72)	AEDC A / SUPERSONIC	A1G		UNASSIGNED
IA244	5*24*77 - 06*01*77	80/ 76	154	5/140C	(72)	LARC 8-FT TRANSONIC PRESSURE	779	2391	PUBLISHED

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PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

AUG 01, 1984

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
CA1	5*30*74 -	06*04*74	40/ 56	50	ET/C-5A	(399)	LOCKHEED (GA) - LOW SPEED			UNASSIGNED
CA2-1	6* 4*74 -	06*10*74	120/ 80	100	4/140A,B/C-5A	(43)	LOCKHEED (GA) - LOW SPEED			UNASSIGNED
CA2-2	6*11*74 -	06*23*74	40/ 40	64	ET/C-5A	(399)	LOCKHEED (GA) - LOW SPEED			UNASSIGNED
CA3	8*15*74 -	08*30*74	120/ 131	194	4/140A,B/747	(43)	UNIV. OF WASH. LOW SPEED	1136	2201	PUBLISHED
CA4	5*28*74 -	06*07*74	64/ 120	100	4/140A,B/747	(43)	UNIV. OF WASH. LOW SPEED	1128		UNASSIGNED
CA5	9*20*74 -	09*30*74	144/ 181	520	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1431	2211	PUBLISHED
CA6	5*20*75 -	06*06*75	200/ 265	509	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1472	2262	PUBLISHED
CA8	8*18*75 -	09*12*75	200/ 324	536	4/140A,B/747	(43)	LARC V/STOL	129	2290	PUBLISHED
CA9	6*25*75 -	07*14*75	320/ 302	85	4/140A,B/747	(47)	THE BOEING CO. - TRANSONIC	1477	2268	PUBLISHED
CA11	2*12*75 -	02*20*75	100/ 116	120	ET/747	(0)	UNIV. OF WASH. LOW SPEED	1146	2236	PUBLISHED
CA13	6* 8*76 -	07*01*76	160/ 193	54	140C(ALT)/747	(45)	ARC 14-FT TRANSONIC	121	2332	PUBLISHED
CA14	11*13*75 -	12*02*75	160/ 236	850	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1496	2307	PUBLISHED
CA15A	10*16*75 -	11*01*75	240/ 239	379	4/140A,B/747	(43)	UNIV. OF WASH. LOW SPEED	1173	2347	PUBLISHED
CA15B	11*19*75 -	11*26*75	75/ 110	93	4/140A,B/747	(43)	UNIV. OF WASH. LOW SPEED	1178	2348	PUBLISHED
CA16	8*23*75 -	09*05*75	72/ 84	60	140A,B/747	(45)	TEXAS A+M 7X10-FT LOW SPEED	7515		UNASSIGNED
CA17	6*21*76 -	07*02*76	152/ 152	261	4/140A,B/747	(43)	UNIV. OF WASH. LOW SPEED	1184	2349	PUBLISHED
CA20	10* 9*74 -	10*15*74	115/ 115	288	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1431	2217	PUBLISHED
CA23A	3*21*75 -	04*17*75	120/ 213	71	140C(MOD)/747	(48)	ARC 14-FT TRANSONIC	085	2243	PUBLISHED
CA23B	5* 1*75 -	07*22*75	160/ 132	46	140C(MOD)/747	(48)	ARC 14-FT TRANSONIC	085	2275	PUBLISHED
CA26	8* 4*75 -	08*15*75	94/ 95	131	140C(MOD)/747	(48)	LTV 4X4-FT SUPERSONIC	559	2273	PUBLISHED
CA92	11*27*73 -	12*04*73	80/ 97	114	089B/747	(2)	THE BOEING CO. - V/STOL	132		UNASSIGNED
CA103	11*26*73 -	11*28*73	24/ 24	45	089B/C-5A	(2)	LOCKHEED (CA) - LOW SPEED	365		UNASSIGNED
CA104	12*13*73 -	01*21*74	160/ 165	208	089B/C-5A	(2)	LOCKHEED (GA) - V/STOL	120		UNASSIGNED

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SA1F	12* 9*72	12*23*72	160/ 144	200	PRR/SRB	(1)	MSFC 14-IN TRANSONIC	554	2012	PUBLISHED
SA2FA	7*24*73	08*07*73	60/ 176	176	SRB	(454)	LARC 8-FT HIGH-TEMP STRUCTURE	655	2088	PUBLISHED
SA2FB	9*24*73	09*28*73	60/ 52	60	SRB	(454)	LARC 8-FT TRANSONIC PRESSURE	662	2088	PUBLISHED
SA3F	2*20*73	03*20*73	160/ 164	261	SRB	(449)	MSFC 14-IN TRANSONIC	565	2025	PUBLISHED
SA5F	5* 3*73	05*08*73	45/ 52	101	SRB	(449)	MSFC 14-IN TRANSONIC	572	2051	PUBLISHED
SA6F	12* 3*73	01*16*74	120/ 208	0	SRB	(454)	LERC 10X10-FT SUPERSONIC	035	2161	PUBLISHED
SA8F	10*18*74	12*10*74	160/ 250	0	SRB	(471)	MSFC 14-IN TRANSONIC	604	2223	PUBLISHED
SA9F	7* 8*74	07*29*74	150/ 256	90	SRB/DROGUE	(0)	LARC 16-FT TRANSONIC	243		UNASSIGNED
SA10F	9*13*73	10*01*73	112/ 128	200	SRB	(449)	MSFC 14-IN TRANSONIC	578	2087	PUBLISHED
SA11FA	4*14*76	04*26*76	200/ 193	0	SRB	(483)	ARC 11-FT TRANSONIC	074	2331	PUBLISHED
SA11FB	4*25*76	05*07*76	120/ 84	0	SRB	(483)	ARC 9X7-FT SUPERSONIC	074	2331	PUBLISHED
SA11FC	3*29*76	04*14*76	120/ 156	0	SRB	(483)	ARC 8X7-FT SUPERSONIC	074	2331	PUBLISHED
SA13F	9*30*74	06*17*75	100/ 100	0	SRB	(461)	MSFC 32-IN LUDWIEG (HIGH RN)	034	2277	PUBLISHED
SA14FA	12*23*75	03*19*76	140/ 144	200	SRB	(449)	MSFC 14-IN TRANSONIC	620	2325	PUBLISHED
SA14FB	1* 6*76	03*11*76	48/ 549	100	SRB	(486)	MSFC 14-IN TRANSONIC	640	2310	PUBLISHED
SA16F	5* 5*76	05*06*76	8/ 8	9	SRB	(486)	AEDC 4-FT TRANSONIC	445	2334	PUBLISHED
SA21F	9*16*76	10*06*76	56/ 120	200	SRB	(486)	MSFC 14-IN TRANSONIC	645	2345	PUBLISHED
SA25F	3* 4*74	03*11*74	40/ 30	16	SRB	(454)	LARC UNITARY PLAN	1087	2150	PUBLISHED
SA26FA	11*19*73	12*11*73	63/ 63	100	SRB	(449)	MSFC 14-IN TRANSONIC	590	2111	PUBLISHED
SA26FB	1*28*74	01*30*74	16/ 13	50	SRB	(449)	MSFC 14-IN TRANSONIC	595	2111	PUBLISHED
SA28F-1	3*17*75	04*11*75	80/ 102	200	SRB	(468)	MSFC 14-IN TRANSONIC	603	2244	PUBLISHED
SA28F-2	3*17*75	04*11*75	40/ 50	160	SRB	(469)	MSFC 14-IN TRANSONIC	603	2244	PUBLISHED
SA29F	8* 8*74	09*18*74	120/ 120	0	SRB FORE BODY	(467)	CALSPAN 32-IN LUDWIEG	033	2207	PUBLISHED
SA30F	3* 3*75	03*13*75	80/ 72	185	SRB	(473)	MSFC 14-IN TRANSONIC	611	2235	PUBLISHED
SA31F	4*27*76	02*01*77	80/ 80	0	SRB	(487)	MSFC 32-IN LUDWIEG (HIGH RN)	039	2369	PUBLISHED

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	REF.	MODEL	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
SA32F	3*22*76 -	04*02*76	150/ 150	94	SRB		(0)	LARC 16-FT TRANSONIC DYNAMIC	275		UNASSIGNED
SA38F	9*23*76 -	09*27*76	32/ 30	0	SRB		(0)	LOCKHEED (GA) - LOW SPEED	190		UNASSIGNED

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TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
TA1F	2*19*74 - 03*05*74	56/ 64	0	ET	(459)	MSFC 14-IN TRANSONIC	583	2145	PUBLISHED
TA2F	4*29*74 - 09*23*74	104/ 104	0	ET	(460)	MSFC 14-IN TRANSONIC	596	2165	PUBLISHED
TA3F	9*27*74 - 10*11*74	64/ 80	0	ET	(470)	MSFC 14-IN TRANSONIC	609	2208	PUBLISHED
TA6F	8* 4*76 - 08*20*76	40/ 80	0	E. INST.	(0)	MSFC 14-IN TRANSONIC	643		UNASSIGNED
TA9F	6* 3*74 - 06*15*74	128/ 144	0	ET	(466)	ARC 3.5-FT HYPERSONIC	196	2181	PUBLISHED

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PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

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TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
MA1	8*25*72	09*06*72	80/ 80	120	JSC 040A ORB.	(95)	LTV 15X20-FT LOW SPEED	407	2004	PUBLISHED
MA2	9*18*72	11*06*72	40/ 80	31	ATP	(1)	LARC 22-IN HELIUM	409	2003	PUBLISHED
MA4	10* 1*72	10*02*72	16/ 16	12	RI ATP ORBITER	(0)	LARC 31-IN CONT-FLOW HYP.	089	2008	PUBLISHED
MA5	9*15*72	09*25*72	80/ 60	30	PRE-ATP/001	(10)	LARC UNITARY PLAN	1002	2001	PUBLISHED
MA6	4* 2*73	04*06*73	120/ 136	4	RI PRR ORB.	(27)	ARC 3.5-FT HYPERSONIC	156		UNASSIGNED
MA7	5*14*73	05*18*73	50/ 50	81	2A/089B	(6)	LARC UNITARY PLAN	1031	2069	PUBLISHED
MA8	12*15*72	01*27*73	40/ 40	40	JSC 040A ORB.	(95)	TEXAS A+M 7X10-FT LOW SPEED	MA8		UNASSIGNED
MA9F	11*29*72	12*07*72	74/ 75	132	ATP	(1)	MSFC 14-IN TRANSONIC	558	2011	PUBLISHED
MA10F	7*23*73	09*12*73	160/ 305	0	OGIVE CYL	(0)	MSFC 14-IN TRANSONIC	575		UNASSIGNED
MA11F	1* 3*74	01*18*74	80/ 102	0	OGIVE CYL	(0)	MSFC 14-IN TRANSONIC	586		UNASSIGNED
MA12F	10* 1*73	02*04*74	80/ 328	0	OGIVE CYL	(0)	MSFC 32-IN LUDWIEG (HIGH RN)	031		UNASSIGNED
MA13	4*15*74	05*03*74	120/ 176	0	GULFSTREAM 2	(0)	ARC 12-FT PRESSURE	028		UNASSIGNED
MA14	4*23*73	05*02*73	8C/ 62	103	2A/089B(CAN)	(95)	LTV 15X20-FT LOW SPEED	422	2283	PUBLISHED
MA16	10* 3*73	10*12*73	40/ 56	106	089B/C-5A	(2)	LOCKHEED (CA) - LOW SPEED	363		UNASSIGNED
MA17	4* 8*74	04*22*74	120/ 152	0	GULFSTREAM 2	(0)	ARC 11-FT TRANSONIC	003		UNASSIGNED
MA18	6* 5*74	06*22*74	200/ 200	254	GULFSTREAM 2	(0)	GRUMMAN - LOW SPEED	324		UNASSIGNED
MA19	8*16*74	09*12*74	120/ 144	36	GULFSTREAM 2	(0)	LARC 16-FT TRANSONIC	295		UNASSIGNED
MA21	8*15*75	09*04*75	80/ 92	50	5/140C	(34)	JPL 20-IN SUPERSONIC	702		UNASSIGNED
MA22	5* 6*75	06*03*75	100/ 168	357	4/140A,B	(32)	LARC 31-IN CONT-FLOW HYP.	118	2267	PUBLISHED
MA24	7* 9*75	08*11*75	24/ 176	200	2A/089B(MOD)	(2)	TEXAS A+M 7X10-FT LOW SPEED	7513		UNASSIGNED
MA28	9*29*76	09*29*76	7/ 1	0	2A/089B	(6)	AEDC A / SUPERSONIC	K8A		UNASSIGNED
MA29	10*14*76	10*14*46	7/ 7	0	SEMISPAN	(0)	AEDC B / HYPERSONIC	K7A	2451	PUBLISHED
MA33A	4*19*82	04*30*82	80/ 144	0	VEH 102	(106)	ARC 11-FT TRANSONIC	510	2507	PUBLISHED
MA33B	5*10*82	05*21*82	40/ 96	0	VEH 102	(106)	ARC 9X7-FT SUPERSONIC	510	2507	PUBLISHED
MA34	3*12*81	03*20*81	40/ 60	0	ADS PROBES	(99)	AEDC 16-FT TRANSONIC	594	2497	IN PROCESS

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
MA35B	12* 2*81 -	12*16*81	40/ 80	0	ADS PROBES	(99)	ARC 9X7-FT SUPERSONIC	513		UNASSIGNED
MA35C	4*19*82 -	04*23*82	40/ 120	0	ADS PROBES	(99)	ARC 8X7-FT SUPERSONIC	513		UNASSIGNED
MA37	11* 2*81 -	11*04*81	40/ 24	100	VEH 102	(106)	LARC UNITARY PLAN	1394		UNASSIGNED

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TEST NO.	TESTING SCHFD.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
FA1	10*10*72 -	11*15*72	416/ 400	200	PRE-ATP/001	(10)	LARC 16-FT TRANSONIC DYNAMIC	210		UNASSIGNED
FA4	1*18*74 -	04*15*74	40/ 182	0	TITAN-3C	(459)	MSFC 14-IN TRANSONIC	587	2142	PUBLISHED
FA6	7*10*74 -	08*19*74	40/ 584	0	OGIVE CYL	(0)	ARC 6X6-FT SUPERSONIC	033		UNASSIGNED
FA7	8*20*74 -	08*28*74	40/ 166	0	OGIVE CYL	(0)	ARC 6X6-FT SUPERSONIC	033		UNASSIGNED
FA8	11* 8*73 -	11*23*74	80/ 80	0	3 ENG. FIRING	(0)	AEDC J	000		UNASSIGNED
FA10	1* 7*74 -	01*14*74	40/ 56	0	JET PLUME SIM.	(0)	MSFC 14-IN TRANSONIC	591		UNASSIGNED
FA11	3*11*74 -	4*08*74	160/ 176	0	CONE-OGIVE-CYL	(0)	MSFC 14-IN TRANSONIC	593		UNASSIGNED
FA12	3* 6*74 -	03*10*74	24/ 24	0	CONE-CYLINDER	(0)	MSFC 14-IN TRANSONIC	597		UNASSIGNED
FA13	8* 7*75 -	09*11*75	160/ 200	0	CONE CYLINDER	(0)	MSFC 14-IN TRANSONIC	612		UNASSIGNED
FA14	1* 9*75 -	07*06*75	60/ 142	0	5/140C	(74)	MSFC 14-IN TRANSONIC	600	2274	PUBLISHED
FA15	1*31*77 -	05*01*77	104/ 320	0	OGIVE CYLINDER	(0)	MSFC 14-IN TRANSONIC	632		UNASSIGNED
FA19	5* 2*77 -	07*05*77	104/ 278	0	ACOUSTICS	(0)	MSFC 14-IN TRANSONIC	630		UNASSIGNED
FA20A	7*18*75 -	07*22*75	24/ 24	0	NOZZLE CALIB.	(0)	MSFC 14-IN TRANSONIC	631		UNASSIGNED
FA20B	10* 3*75 -	10*13*75	60/ 52	0	TRIPLE BODY	(0)	AEDC 4-FT TRANSONIC	409		UNASSIGNED
FA21A	10*23*75 -	10*31*75	24/ 44	0	NOZZLE CALIB.	(0)	MSFC 14-IN TRANSONIC	633		UNASSIGNED
FA21B	2*17*76 -	04*21*76	0/ 368	0	ROCKET MOTOR	(0)	MSFC 32-IN LUDWIEG (HIGH RN)	038		UNASSIGNED
FA22A	7* 9*75 -	07*17*75	12/ 56	0	NOZZLE CALIB.	(0)	MSFC 14-IN TRANSONIC	628		UNASSIGNED
FA22B	7*21*75 -	07*25*75	30/ 35	0	OGIVE CYLINDER	(0)	AEDC 4-FT TRANSONIC	390		UNASSIGNED
FA23A	11* 3*75 -	11*06*75	60/ 28	0	NOZZLE CALIB.	(0)	MSFC 14-IN TRANSONIC	627		UNASSIGNED
FA23B	12* 1*75 -	12*12*75	60/ 72	0	TRIPLE BODY	(0)	ARC 9X7-FT SUPERSONIC	103		UNASSIGNED
FA23C	7*24*76 -	08*04*76	60/ 44	0	TRIPLE BODY	(0)	ARC 11-FT TRANSONIC	103		UNASSIGNED
FA25	4*15*78 -	08*01*78	200/ 294	0	5/140C	(74)	MSFC 14-IN TRANSONIC	652	2437	PUBLISHED
FA26	5* 1*78 -	06*01*78	80/ 80	0	5/140C	(74)	MSFC 14-IN TRANSONIC	653		UNASSIGNED
FA27	3*14*79 -	05*16*79	150/ 160	0	5/140C	(74)	MSFC 14-IN TRANSONIC	655	2460	IN PROCESS
FA28	8* 1*79 -	09*01*79	200/ 0	0	5/140C	(74)	MSFC 14-IN TRANSONIC	656	2474	PUBLISHED

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	STATUS
FA29	4* 1*79 -	05*00*79	400/ 0	0	5/140C	(74)	MSFC 14-IN TRANSONIC	657		UNASSIGNED

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TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
LA1	11*19*72	12*19*72	60/ 84	73	ATP	(6)	LARC 8-FT TRANSONIC PRESSURE	626	2002	PUBLISHED
LA2	10* 6*72	12*07*72	120/ 136	24	L/0-100 ORB.	(0)	LARC 22-IN HELIUM	411	2023	PUBLISHED
LA3	8*23*72	11*16*72	40/ 46	19	L/0-100 ORB.	(0)	LARC 31-IN CONT-FLOW HYP.	085	2031	PUBLISHED
LA4A	11* 2*72	12*06*72	80/ 75	37	L/0-100 ORB.	(0)	LARC UNITARY PLAN	1014	2033	PUBLISHED
LA4B	10*25*72	11*01*72	80/ 60	32	L/0-100 ORB.	(0)	LARC UNITARY PLAN	995	2033	PUBLISHED
LA4C	2*19*73	02*23*73	80/ 50	43	L/0-100 ORB.	(0)	LARC UNITARY PLAN	995	2033	PUBLISHED
LA6	4*12*73	04*18*73	72/ 72	108	089B, 139 NOSE	(0)	LARC 8-FT TRANSONIC PRESSURE	643	2040	PUBLISHED
LA8A	4*18*73	04*24*73	50/ 45	58	089B, 139 NOSE	(0)	LARC UNITARY PLAN	1023	2054	PUBLISHED
LA8B	5* 7*73	05*15*73	50/ 70	50	089B, 139 NOSE	(0)	LARC UNITARY PLAN	1034	2054	PUBLISHED
LA8C	7* [illegible]*73	07*06*73	50/ 30	14	089B, 139 NOSE	(0)	LARC UNITARY PLAN	1039	2054	PUBLISHED
LA8D	7*10*73	07*13*73	50/ 42	37	089B, 139 NOSE	(0)	LARC UNITARY PLAN	1040	2090	PUBLISHED
LA9A	4*26*73	05*07*73	160/ 96	65	089B,139 NOSE	(0)	LARC LOW TURBULANCE PRESSURE	130	2056	PUBLISHED
LA9B	5*23*73	05*31*73	140/ 32	22	089B,139 NOSE	(0)	LARC LOW TURBULANCE PRESSURE	135	2056	PUBLISHED
LA9C	10*31*73	11*02*73	140/ 32	28	089B,139 NOSE	(0)	LARC LOW TURBULANCE PRESSURE	148	2056	PUBLISHED
LA11	7*11*73	07*20*73	24/ 58	85	089B, 139 NOSE	(0)	LARC 31-IN CONT-FLOW HYP.	096	2066	PUBLISHED
LA12A	9* 4*73	09*17*73	40/ 80	15	089B, 139 NOSE	(0)	LARC 22-IN HELIUM	418		UNASSIGNED
LA12B	9*18*73	01*17*74	272/ 272	56	089B,139 NOSE	(0)	LARC 22-IN HELIUM	419		UNASSIGNED
LA13A	8*17*73	08*28*73	40/ 64	15	089B, 139 NOSE	(0)	LARC 31-IN CONT-FLOW HYP.	099	2135	CANCEL
LA13B	10* 9*73	10*15*73	40/ 40	16	089B, 139 NOSE	(0)	LARC 31-IN CONT-FLOW HYP.	099	2135	CANCEL
LA13C	11*14*73	11*16*73	40/ 24	31	089B, 139 NOSE	(0)	LARC 31-IN CONT-FLOW HYP.	099	2135	CANCEL
LA14A	8*17*73	08*28*73	100/ 80	20	089B, 139 NOSE	(202)	LARC UNITARY PLAN	1046	2106	PUBLISHED
LA14B	8* 6*73	08*16*73	100/ 90	47	089B, 139 NOSE	(202)	LARC UNITARY PLAN	1049	2106	PUBLISHED
LA14C	9* 5*73	09*10*73	100/ 40	45	089B, 139 NOSE	(202)	LARC UNITARY PLAN	1055	2106	PUBLISHED
LA14D	12* 5*73	12*07*73	100/ 30	9	089B, 139 NOSE	(202)	LARC UNITARY PLAN	1058	2106	PUBLISHED
LA15	8* 3*73	09*24*73	120/ 240	69	089B, 139 NOSE	(0)	LARC 20-IN HYPERSONIC (M=6)	6441	2079	PUBLISHED

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TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
LA16	6*26*72 - 08*23*72	60/ 64	72	HRSI TILE	(0)	LARC MACH 8 VARIABLE DENSITY	624	2043	PUBLISHED
LA17	5* 4*73 - 05*14*73	80/ 96	102	L/O-100 ORB	(0)	LARC 8-FT TRANSONIC PRESSURE	648	2046	PUBLISHED
LA20A	6*22*73 - 07*06*73	160/ 160	81	089B,139NOSE	(202)	LARC 8-FT TRANSONIC PRESSURE	653	2107	PUBLISHED
LA20B	8*21*74 - 08*26*74	160/ 54	20	089B,139NOSE	(202)	LARC 8-FT TRANSONIC PRESSURE	692	2107	PUBLISHED
LA20C	8*29*73 - 08*31*73	160/ 44	50	089B,139NOSE	(202)	LARC 8-FT TRANSONIC PRESSURE	658	2107	PUBLISHED
LA21A	8*19*74 - 08*30*74	160/ 144	55	089B, 139 NOSE	(0)	LARC LOW TURBULANCE PRESSURE	202		UNASSIGNED
LA21B	1*29*75 - 02*05*75	80/ 88	37	089B, 139 NOSE	(0)	LARC LOW TURBULANCE PRESSURE	206		UNASSIGNED
LA22	6*19*72 - 06*30*72	160/ 152	31	JSC 049	(0)	LARC 22-IN HELIUM	405	2034	PUBLISHED
LA23	7*31*73 - 08*03*73	48/ 32	15	L/O-100 ORB.	(0)	LARC LOW TURBULANCE PRESSURE	141	2070	PUBLISHED
LA24A	11* 9*73 - 11*12*73	40/ 20	6	089B, 139 NOSE	(202)	LARC UNITARY PLAN	1065		UNASSIGNED
LA24B	1* 2*74 - 01*07*74	40/ 34	20	089B, 139 NOSE	(202)	LARC UNITARY PLAN	1065		UNASSIGNED
LA25	8*30*73 - 09*07*73	40/ 48	126	3/139B	(32)	LARC 31-IN CONT-FLOW HYP.	100	2126	CANCEL
LA28	6*17*74 - 06*20*74	40/ 40	31	140A,B ORB	(0)	LTV 4X4-FT SUPERSONIC	498	2280	PUBLISHED
LA31	8* 9*73 - 08*16*73	48/ 72	28	LARC ORB	(0)	LARC 31-IN CONT-FLOW HYP.	098	2047	PUBLISHED
LA32A	7*25*73 - 08*03*73	180/ 64	16	F.S. TILE ARRAY	(0)	LARC 31-IN CONT-FLOW HYP.	097	2168	PUBLISHED
LA32B	11*28*73 - 12*03*73	180/ 120	43	F.S. TILE ARRAY	(0)	LARC 31-IN CONT-FLOW HYP.	097	2168	PUBLISHED
LA33	11*19*73 - 11*26*73	40/ 48	26	089B, 139 NOSE	(0)	LARC 31-IN CONT-FLOW HYP.	103		UNASSIGNED
LA34	1*17*74 - 01*31*74	40/ 112	55	F.S. TILE ARRAY	(0)	LARC 31-IN CONT-FLOW HYP.	105	2328	PUBLISHED
LA35	11*12*73 - 11*13*73	16/ 20	19	3/139B	(32)	LARC 31-IN CONT-FLOW HYP.	102	2127	PUBLISHED
LA36A	11* 5*73 - 11*11*73	75/ 75	22	140A,B	(42)	LARC 16-FT TRANSONIC	149		UNASSIGNED
LA36B	6* 3*75 - 06*05*75	75/ 27	41	140A,B	(32)	LARC LOW TURBULANCE PRESSURE	214	2292	PUBLISHED
LA38A	12*14*73 - 12*21*73	56/ 56	59	140A,B	(0)	LARC 8-FT TRANSONIC PRESSURE	669	2121	CANCEL
LA38B	3*27*74 - 03*29*74	48/ 48	37	140A,B	(0)	LARC 8-FT TRANSONIC PRESSURE	676	2239	PUBLISHED
LA39A	12*26*73 - 12*28*73	50/ 24	9	140A,B	(0)	LARC UNITARY PLAN	1075	2188	PUBLISHED
LA39B	2*11*74 - 02*15*74	50/ 50	36	140A,B	(0)	LARC UNITARY PLAN	1075	2188	PUBLISHED

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TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
LA39C	4* 1*74 - 04*08*74	50/ 80	26	140A,B	(0)	LARC UNITARY PLAN	1075	2188	PUBLISHED
LA40	5*13*74 - 06*07*74	40/ 40	25	139B	(0)	LARC 22-IN HELIUM	7426	2176	PUBLISHED
LA42A	6*25*74 - 06*25*74	16/ 8	3	089B	(0)	AEDC B / HYPERSONIC	550	2132	PUBLISHED
LA42B	7*27*74 - 07*27*74	16/ 12	7	089B	(0)	AEDC B / HYPERSONIC	48A	2132	PUBLISHED
LA43A	3* 4*74 - 03*22*74	50/ 30	42	4/140A,B	(0)	LARC UNITARY PLAN	1074	2199	PUBLISHED
LA43B	3*18*74 - 03*27*74	50/ 70	28	4/140A,B	(0)	LARC UNITARY PLAN	1093	2199	PUBLISHED
LA44	4* 2*74 - 04*09*74	160/ 96	54	4/140A,B	(0)	LARC 8-FT TRANSONIC PRESSURE	677	2200	PUBLISHED
LA46A	9*13*74 - 09*24*74	96/ 96	61	140A,B ORB	(0)	LARC UNITARY PLAN	1092	2228	PUBLISHED
LA46B	9*24*74 - 10*10*74	88/ 88	51	140A,B ORB	(0)	LARC UNITARY PLAN	1117	2228	PUBLISHED
LA47A	1* 2*74 - 01*09*74	40/ 120	43	140A/B ORB	(0)	LARC 31-IN CONT-FLOW HYP.	104	2191	PUBLISHED
LA47B	6*10*74 - 06*24*74	40/ 88	35	140A/B ORB	(0)	LARC 31-IN CONT-FLOW HYP.	104	2191	PUBLISHED
LA47C	7* 8*74 - 07*10*74	40/ 16	18	140A/B ORB	(0)	LARC 31-IN CONT-FLOW HYP.	104	2191	PUBLISHED
LA48	4*10*74 - 04*15*74	48/ 48	99	089B-MOD NOSE	(0)	LARC 8-FT TRANSONIC PRESSURE	680	2184	PUBLISHED
LA49A	4*24*74 - 04*26*74	20/ 30	37	089B-MOD NOSE	(0)	LARC UNITARY PLAN	1101	2182	PUBLISHED
LA49B	7*15*74 - 07*17*74	20/ 25	105	089B-MOD NOSE	(0)	LARC UNITARY PLAN	1111	2182	PUBLISHED
LA51	5*24*74 - 05*31*74	80/ 72	140	140A,B	(0)	LARC 8-FT TRANSONIC PRESSURE	684	2183	PUBLISHED
LA52	8*26*74 - 08*30*74	72/ 40	38	140A,B	(0)	LARC 20-IN HYPERSONIC (M=6)	6458	2220	PUBLISHED
LA53A	8*12*74 - 08*14*74	80/ 32	3	5/140C	(0)	LARC 20-IN FREON	220	2213	IN PROCESS
LA53B	1*12*77 - 01*18*77	80/ 72	16	5/140C	(0)	LARC 22-IN HELIUM	306	2213	IN PROCESS
LA54	8*14*74 - 08*19*74	28/ 28	5	140C ORB.	(0)	LARC 20-IN HYPERSONIC (M=6)	6456	2213	IN PROCESS
LA56	11*11*74 - 11*22*74	160/ 176	147	VEH. 5	(0)	LARC 8-FT TRANSONIC PRESSURE	699	2224	PUBLISHED
LA57A	10* 2*74 - 10*24*74	84/ 144	58	140A,B	(0)	LARC 31-IN CONT-FLOW HYP.	114	2454	PUBLISHED
LA57B	6* 4*75 - 06*06*75	84/ 24	10	140A,B	(0)	LARC 31-IN CONT-FLOW HYP.	114	2454	PUBLISHED
LA58	9*30*74 - 10*04*74	49/ 80	72	140A,B	(42)	LTV 4X4-FT SUPERSONIC	512	2215	PUBLISHED
LA59	12*20*74 - 01*07*75	96/ 96	146	4/140A,B	(72)	LARC 8-FT TRANSONIC PRESSURE	703	2233	PUBLISHED

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TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	ST ATUS
LA61A	8*25*75	09*10*75	40/ 40	138	140C/REMOTE ELE	(44)	LARC LOW TURBULANCE PRESSURE	219	2278	CANCEL
LA61B	1* 5*76	01*14*76	40/ 96	81	140C/REMOTE ELE	(44)	LARC LOW TURBULANCE PRESSURE	228	2300	PUBLISHED
LA62	5*14*75	05*23*75	40/ 80	301	140C/REMOTE ELE	(44)	LARC 8-FT TRANSONIC PRESSURE	717	2264	PUBLISHED
LA63A	7*18*75	07*18*75	40/ 14	63	140C/REMOTE ELE	(44)	LARC UNITARY PLAN	1118	2270	PUBLISHED
LA63B	9*12*75	09*17*75	40/ 38	191	140C/REMOTE ELE	(44)	LARC UNITARY PLAN	1151	2279	PUBLISHED
LA66	10*20*75	10*24*75	80/ 80	26	140C/REMOTE ELE	(44)	ARC 12-FT PRESSURE	135	2281	PUBLISHED
LA67	6*20*75	07*02*75	40/ 120	131	140C/REMOTE ELE	(44)	LTV 4X4-FT SUPERSONIC	552	2266	PUBLISHED
LA68	2*26*75	03*20*75	120/ 120	26	140C ORB	(0)	LARC 22-IN HELIUM	439	2256	IN PROCESS
LA69	4*24*75	04*29*75	64/ 64	98	5/140C	(72)	LARC 8-FT TRANSONIC PRESSURE	714	2257	PUBLISHED
LA70	7*28*75	08*06*75	38/ 60	299	140C/REMOTE ELE	(44)	CALSPAN 8-FT TRANSONIC	103	2269	PUBLISHED
LA71B	7*21*75	07*31*75	48/ 48	30	4/140A,B	(69)	LARC UNITARY PLAN	1147	2271	PUBLISHED
LA71A	10*17*75	10*22*75	48/ 64	15	4/140A,B	(69)	LARC UNITARY PLAN	1132	2271	PUBLISHED
LA72	3*26*76	03*31*76	72/ 72	30	4/140A,B	(69)	LARC 8-FT TRANSONIC PRESSURE	740	2309	PUBLISHED
LA73A	12*18*75	12*30*75	82/ 82	14	4/140A,B	(69)	LARC LOW TURBULANCE PRESSURE	227	2298	PUBLISHED
LA73B	12*10*76	12*13*75	16/ 16	6	4/140A,B	(69)	LARC LOW TURBULANCE PRESSURE	238	2298	PUBLISHED
LA75	4* 6*76	04*16*76	90/ 90	283	140C/REMOTE ELE	(44)	LARC UNITARY PLAN	1173	2318	PUBLISHED
LA76	2*25*76	03*06*76	48/ 128	141	140C/REMOTE ELE	(44)	LTV 4X4-FT SUPERSONIC	573	2305	PUBLISHED
LA77	7* 9*76	07*24*76	120/ 151	521	140C/REMOTE ELE	(44)	ARC 11-FT TRANSONIC	200	2344	PUBLISHED
LA78	1*15*76	01*28*76	16/ 16	4	089B	(13)	LARC 4-FT HYPERSONIC	267	2311	PUBLISHED
LA79	11*28*75	12*11*75	64/ 64	8	140C	(0)	NSWC HYPERVELOCITY LAB (#8A)	1275	2291	IN PROCESS
LA80	10* 6*75	11*07*75	156/ 156	83	140C/747	(88)	LARC 7X10-FT HIGH SPEED	999	2299	PUBLISHED
LA81	1*14*76	01*23*76	40/ 120	54	ORB/TC (ALT)	(0)	LARC LOW TURBULANCE PRESSURE	229	2296	PUBLISHED
LA82	8* 8*76	08*19*76	30/ 32	66	SUPPORT TARES	(202)	CALSPAN 8-FT TRANSONIC	111	2374	PUBLISHED
LA85	4* 7*76	05*24*76	88/ 88	64	140C	(13)	LARC 22-IN HELIUM	445	2343	PUBLISHED
LA87	8*26*75	08*29*75	36/ 36	4	089B	(13)	LARC 4-FT HYPERSONIC	446	2311	PUBLISHED

TEST NO.	TESTING SCHED.	COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
LA88	5*21*75 -	05*21*75	16/ 16	6	089B	(13)	LARC 20-IN HYPERSONIC (M=6)	6468	2311	PUBLISHED
LA89	10* 4*76 -	10*18*76	160/ 208	220	OV101 (ALT)	(201)	ARC 11-FT TRANSONIC	213	2353	PUBLISHED
LA91	9* 3*76 -	09*15*76	80/ 104	214	140C/REMOTE ELE	(44)	LARC 8-FT TRANSONIC PRESSURE	758	2352	PUBLISHED
LA92	11*11*76 -	11*19*76	80/ 152	67	OV101	(201)	LARC 8-FT TRANSONIC PRESSURE	764	2362	IN PROCESS
LA93	3*16*77 -	04*21*77	80/ 56	34	NOSE CONE	(0)	LARC 31-IN CONT-FLOW HYP.	130	2383	IN PROCESS
LA95	6* 6*77 -	10*19*77	160/ 160	14	NOSE CONE	(0)	LARC 20-IN FREON	330		UNASSIGNED
LA97A	4*18*77 -	08*15*77	80/ 400	116	L.E.VORTEX	(0)	MSFC 32-IN LUDWIEG (HIGH RN)	041		UNASSIGNED
LA97B	10* 1*78 -	06*01*79	80/ 400	0	L.E.VORTEX	(0)	MSFC 32-IN LUDWIEG (HIGH RN)	041		UNASSIGNED
LA98	1*28*77 -	02*02*77	32/ 64	42	L.E. VORTEX	(0)	LARC LOW TURBULANCE PRESSURE	240		UNASSIGNED
LA99	2*17*77 -	02*28*77	104/ 104	147	TAILCONE	(201)	LARC 8-FT TRANSONIC PRESSURE	769	2373	PUBLISHED
LA100A	12*15*76 -	12*17*76	14/ 14	4	GENERIC	(0)	LARC LOW TURBULANCE PRESSURE	239		UNASSIGNED
LA100B	2* 2*77 -	02*07*77	25/ 25	24	GENERIC	(0)	LARC LOW TURBULANCE PRESSURE	241		UNASSIGNED
LA101	5*18*77 -	05*24*77	55/ 55	200	140C/REMOTE ELE	(44)	LARC UNITARY PLAN	1194	2390	PUBLISHED
LA102	12* 7*78 -	01*04*79	40/ 120	29	STING EFFECT	(0)	LARC 22-IN HELIUM	463		UNASSIGNED
LA103	3*25*77 -	04*04*77	80/ 75	88	SUPPORT TARES	(0)	CALSPAN 8-FT TRANSONIC	113	2374	PUBLISHED
LA110	8* 8*77 -	08*10*77	30/ 30	60	140C SILTS	(44)	LARC UNITARY PLAN	1212	2396	PUBLISHED
LA111	8* 3*77 -	08*05*77	95/ 40	95	140C SILTS	(44)	LARC 8-FT TRANSONIC PRESSURE	786	2395	PUBLISHED
LA112	2* 3*77 -	02*05*77	24/ 24	0	5/140C	(0)	LARC 20-IN HYPERSONIC (M=6)	6502		UNASSIGNED
LA113	8* 5*77 -	09*08*77	32/ 28	17	5/140C	(72)	LARC 8-FT TRANSONIC PRESSURE	787	2397	PUBLISHED
LA114	8*23*77 -	08*31*77	30/ 60	70	140C SILTS	(44)	LARC UNITARY PLAN	1217	2399	PUBLISHED
LA115	2* 1*78 -	02*06*78	45/ 45	175	140C/REMOTE ELE	(44)	LARC 8-FT TRANSONIC PRESSURE	803	2409	PUBLISHED
LA116	2* 6*78 -	02*06*78	32/ 32	0	140C	(201)	LARC 8-FT TRANSONIC PRESSURE	804	2411	CANCEL
LA124	6* 7*77 -	06*10*77	40/ 40	19	5/140C	(74)	LARC UNITARY PLAN	1207	2426	PUBLISHED
LA125	7* 3*78 -	07*05*78	16/ 48	41	VEH. 102	(105)	LARC UNITARY PLAN	1243	2432	PUBLISHED
LA131	1* 8*80 -	02*01*80	60/ 144	624	VEH 102	(106)	LARC UNITARY PLAN	1299	2478	PUBLISHED

TEST NO.	TESTING SCHED.	COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
LA132	10*11*79 -	11*01*79	40/ 80	18	VEH 102	(89)	LARC 16-FT TRANSONIC	341	2471	PUBLISHED
LA140	12*26*79 -	01*03*80	80/ 80	17	VEH. 102	(105)	LARC 16-FT TRANSONIC	342	2475	PUBLISHED
LA141A	1*12*80 -	02*01*80	80/ 148	0	VEH 102	(74)	LARC 20-IN HYPERSONIC (M=6)	6546	2477	PUBLISHED
LA141B	3*18*80 -	05*01*80	80/ 200	0	VEH 102	(74)	LARC 20-IN HYPERSONIC (M=6)	6546	2477	PUBLISHED
LA141C	6*22*80 -	07*01*80	80/ 10	0	VEH 102	(72)	LARC 20-IN HYPERSONIC (M=6)	6550		UNASSIGNED
LA142	2* 1*80 -	03*01*80	80/ 80	0	VEH 102	(74)	LARC 20-IN FREON	390		UNASSIGNED
LA143	12*21*79 -	01*08*80	80/ 88	0	VEH 102	(106)	LARC 8-FT TRANSONIC PRESSURE	865		UNASSIGNED
LA144	7*28*80 -	08*01*80	80/ 138	198	VEH 102	(106)	LTV 4X4-FT SUPERSONIC	742	2484	PUBLISHED
LA145A	9*28*81 -	10*08*81	80/ 90	32	140C	(203)	LARC UNITARY PLAN	1390	2336	PUBLISHED
LA145B	9*11*81 -	09*17*81	80/ 50	37	140C	(203)	LARC UNITARY PLAN	1345	2336	PUBLISHED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

AUG 01,1984

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
AA1A	2*12*74	02*25*74	64/ 64	28	4/140A,B	(49)	ARC 3.5-FT HYPERSONIC	186		UNASSIGNED
AA1B	4* 4*74	04*06*74	64/ 64	13	4/140A,B	(49)	ARC 3.5-FT HYPERSONIC	186		UNASSIGNED
AA2	10*30*75	11*01*75	40/ 40	0	TUNNEL CALIB.	(0)	ARC 40X80-FT SUBSONIC	471		UNASSIGNED
AA3A	1*10*71	01*31*71	300/ 300	0	PRE-ATP	(0)	ARC 11-FT TRANSONIC	608	2255	PUBLISHED
AA3B	2* 1*71	02*15*71	100/ 100	0	PRE-ATP	(0)	ARC 9X7-FT SUPERSONIC	608	2255	PUBLISHED

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APPENDIX A

TABLE A2 - WIND TUNNEL TESTING BY TEST NUMBER - HEATING TESTS

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

AUG 01, 1984

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
OH1A-1	9*19*72	09*26*72	10/ 10	130	PRE-ATP/001	(3)	LARC MACH 8 VARIABLE DENSITY	3234		UNASSIGNED
OH1A-2	9*19*72	09*26*72	10/ 100	120	PRE-ATP/001	(4)	LARC MACH 8 VARIABLE DENSITY	3234		UNASSIGNED
OH1A-3	9*19*72	09*26*72	10/ 100	120	PRE-ATP/001	(5)	LARC MACH 8 VARIABLE DENSITY	3234		UNASSIGNED
OH1A-4	9*19*72	09*26*72	10/ 100	120	PRE-ATP/001	(38)	LARC MACH 8 VARIABLE DENSITY	3234		UNASSIGNED
OH1B	11* 6*72	11*08*72	40/ 24	35	PRE-ATP/001	(4)	LARC MACH 8 VARIABLE DENSITY	3283		UNASSIGNED
OH2	4*18*73	06*01*73	40/ 144	81	TPS TILES	(15)	ARC 3.5-FT HYPERSONIC	158	2035	PUBLISHED
OH3A	6*28*73	06*30*73	40/ 16	36	3/139B	(21)	AEDC B / HYPERSONIC	288	2100	PUBLISHED
OH3B	7* 9*73	07*11*73	40/ 23	147	3/139B	(21)	AEDC B / HYPERSONIC	289	2100	PUBLISHED
OH4A	11*12*73	12*05*73	20/ 20	57	3/139	(29)	AEDC B / HYPERSONIC	352	2154	PUBLISHED
OH4B	9*29*73	10*04*73	48/ 38	224	3/139	(22)	AEDC B / HYPERSONIC	352	2099	PUBLISHED
OH4C	9*26*73	09*26*73	8/ 8	60	3/139B	(21)	AEDC B / HYPERSONIC	352	2225	PUBLISHED
OH6	.* 6*74	02*11*74	48/ 56	39	3/139	(22)	ARC 3.5-FT HYPERSONIC	183	2151	PUBLISHED
OH8F	5*15*74	07*16*74	340/ 334	66	2A/089B	(25)	MSFC IMPULSE BASE FLOW FAC.	027	2382	PUBLISHED
OH9	9*13*73	09*21*73	16/ 16	61	3/139	(29)	AEDC B / HYPERSONIC	353	2251	PUBLISHED
OH10	8*17*73	09*04*73	96/ 104	35	3/139	(26)	ARC 3.5-FT HYPERSONIC	171	2085	PUBLISHED
OH11	10*24*73	11*01*73	40/ 37	23	3/139	(29)	AEDC F / HYPERSONIC	VA354	2141	PUBLISHED
OH12	10*29*73	12*13*73	80/ 145	32	3/139	(37)	CALSPAN HYPERSONIC SHOCK	100	2164	PUE ISHED
OH13	6*13*73	06*13*73	8/ 8	18	2A/089B	(41)	LARC MACH 8 VARIABLE DENSITY	644	2096	PUBLISHED
OH14	10*17*73	10*18*73	16/ 16	29	3A/139B	(50)	LARC MACH 8 VARIABLE DENSITY	648	2117	PUBLISHED
OH15	9*12*73	09*20*73	64/ 96	32	FLAT PLATE	(53)	ARC 3.5-FT HYPERSONIC	173	2385	PUBLISHED
OH25A	8*21*74	08*22*74	12/ 12	82	3/139B	(21)	AEDC B / HYPERSONIC	83A	2252	PUBLISHED
OH25B	1*30*75	02*03*75	24/ 23	153	5/140C	(56)	AEDC B / HYPERSONIC	83A	2366	PUBLISHED
OH26	7*22*74	07*29*74	80/ 96	56	4/140B	(22)	ARC 3.5-FT HYPERSONIC	199	2193	PUBLISHED
OH38	6*21*74	07*18*74	160/ 320	91	4/140B	(61)	ARC 3.5-FT HYPERSONIC	198	2171	PUBLISHED
OH39A	11*21*74	11*28*74	84/ 59	622	5/140C	(60)	AEDC B / HYPERSONIC	74A	2241	PUBLISHED

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	ST ATUS
OH39B	1* 8*75	- 01*09*75	12/ 13	80	5/140C	(60)	AEDC B / HYPERSONIC	74A	2241	PUBLISHED
OH40	1*30*73	- 02*05*73	36/ 40	52	2A/089B	(31)	LARC MACH 8 VARIABLE DENSITY	3619	2049	PUBLISHED
OH41A	3*19*73	- 03*28*73	40/ 64	78	2A/089B	(33)	LARC MACH 8 VARIABLE DENSITY	3778	2075	PUBLISHED
OH41B	5* 8*73	- 05*10*73	40/ 24	20	2A/089B	(38)	LARC MACH 8 VARIABLE DENSITY	4060/72	2076	PUBLISHED
OH42A	5*14*73	- 05*16*73	20/ 20	20	3/139,139A	(46)	LARC MACH 8 VARIABLE DENSITY	4080	2101	PUBLISHED
OH42B	5*25*73	- 06*01*73	40/ 48	64	3/139, 139A	(46)	LARC MACH 8 VARIABLE DENSITY	4080	2101	PUBLISHED
OH42C	6*14*73	- 06*15*73	20/ 16	26	3/139A, W/CAN	(46)	LARC MACH 8 VARIABLE DENSITY	4080	2101	PUBLISHED
OH43	12* 2*73	- 12*21*73	160/ 128	92	TPS TILES	(15)	ARC 3.5-FT HYPERSONIC	182	2250	PUBLISHED
OH44	10*24*73	- 10*30*73	180/ 80	46	FLAT PLATE	(53)	ARC 3.5-FT HYPERSONIC	177	2386	PUBLISHED
OH45	11* 2*73	- 11*09*73	40/ 46	22	3A/139B	(50)	LARC 20-IN FREON	121	2109	PUBLISHED
OH46	11*12*73	- 12*07*73	40/ 72	100	4/140B	(90)	LARC MACH 8 VARIABLE DENSITY	4556	2350	PUBLISHED
OH49A	4* 3*74	- 04*06*74	216/ 17	87	3/139B	(22)	AEDC B / HYPERSONIC	525	2355	PUBLISHED
OH49B	7* 2*74	- 07*12*74	72/ 67	454	4/140B	(22)	AEDC B / HYPERSONIC	57A	2222	PUBLISHED
OH50A	3*29*74	- 04*11*74	8/ 16	66	5/140C	(82)	AEDC B / HYPERSONIC	526	2285	PUBLISHED
OH50B	7*12*74	- 07*17*74	36/ 27	220	5/140C	(83)	AEDC B / HYPERSONIC	58A	2358	PUBLISHED
OH51-1	6*26*74	- 07*03*74	24/ 30	50	3/139B	(64)	LARC 31-IN CONT-FLOW HYP.	112	2368	PUBLISHED
OH51-2	6*26*74	- 07*03*74	12/ 180	280	3/139B	(46)	LARC 31-IN CONT-FLOW HYP.	112	2368	PUBLISHED
OH51-3	6*26*74	- 07*03*74	12/ 100	100	4/140B	(90)	LARC 31-IN CONT-FLOW HYP.	112	2368	PUBLISHED
OH52	5* 6*74	- 05*15*74	16/ 16	32	3/139B	(29)	AEDC B / HYPERSONIC	524	2330	PUBLISHED
OH53A	4* 7*76	- 04*13*76	40/ 40	39	5/140C	(83)	ARC 3.5-FT HYPERSONIC	216	2317	PUBLISHED
OH53B	4*14*76	- 04*23*76	40/ 80	23	5/140C	(82)	ARC 3.5-FT HYPERSONIC	216	2317	PUBLISHED
OH54A	10* 4*74	- 10* 8*74	36/ 32	117	5/140C	(82)	AEDC B / HYPERSONIC	82A	2301	PUBLISHED
OH54B	7*21*75	- 07*25*75	48/ 52	124	5/140C	(82)	AEDC B / HYPERSONIC	82A	2342	PUBLISHED
OH54C	8*26*75	- 09*02*75	48/ 48	120	5/140C	(82)	AEDC B / HYPERSONIC	82A	2342	PUBLISHED
OH56	12* 6*77	- 12*10*77	48/ 36	255	WING TIP SEAL	(91)	AEDC B / HYPERSONIC	R3A	2410	PUBLISHED

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
0H57A	10* 6*76	- 20*06*76	13/ 11	40	140C	(92)	AEDC B / HYPERSONIC	K3A	2367	PUBLISHED
0H57B	12* 4*76	- 12*05*76	26/ 34	14	140C	(92)	AEDC B / HYPERSONIC	K3A	2367	PUBLISHED
0H58	3*24*78	- 04*21*78	120/ 168	58	ELEV/ELEV SEAL	(93)	ARC 3.5-FT HYPERSONIC	235	2417	PUBLISHED
0H60	5*12*75	- 05*12*75	12/ 12	139	5/140C	(83)	AEDC B / HYPERSONIC	B7A	2356	PUBLISHED
0H64	4*14*75	- 06*20*75	200/ 450	200	2A/089B	(25)	LERC SPACE POWER FACILITY	0H64	2288	PUBLISHED
0H66	8*30*76	- 10*17*76	120/ 120	30	5/140C	(66)	CALSPAN HYPERSONIC SHOCK	131	2359	PUBLISHED
0H69	11*14*75	- 12*11*75	84/ 87	246	5/140C	(82)	AEDC B / HYPERSONIC	E9A	2321	PUBLISHED
0H74	6* 3*75	- 06*12*75	12/ 12	0	5/140C	(56)	AEDC B / HYPERSONIC	B7A	2263	PUBLISHED
0H75	9* 2*75	- 09*03*75	14/ 13	44	5/140C	(82)	AEDC B / HYPERSONIC	E3A	2303	PUBLISHED
0H78	7* 2*76	- 11*24*76	480/ 1	0	5/140C	(65)	JSC VAC. CHAMBER A	56-A-76	2371	PUBLISHED
0H79	6* 1*78	- 08*24*78	288/ 288	0	5/140C	(65)	JSC VAC. CHAMBER A	61-A-78	2443	PUBLISHED
0H84A-1	4*20*77	- 04*21*77	20/ 16	81	5/140C	(60)	AEDC B / HYPERSONIC	R4A	2388	PUBLISHED
0H84A-2	4*20*77	- 04*21*77	5/ 9	16	5/140C	(83)	AEDC B / HYPERSONIC	R4A	2388	PUBLISHED
0H84B	5* 0*79	- 06*00*79	72/ 72	0	5/140C	(60)	AEDC B / HYPERSONIC	B67	2464	PUBLISHED
0H84C	6*15*79	- 06*28*79	80/ 80	0	5/140C	(60)	ARC 3.5-FT HYPERSONIC	246	2468	PUBLISHED
0H90	3* 2*78	- 03*11*78	48/ 64	162	ELEV/ELEV	(94)	AEDC B / HYPERSONIC	P4A	2451	PUBLISHED
0H98A	6*17*76	- 06*23*76	43/ 44	284	5/140C	(60)	AEDC B / HYPERSONIC	J7A	2340	PUBLISHED
0H98B	7*26*76	- 07*26*76	20/ 13	98	5/140C	(60)	AEDC B / HYPERSONIC	J74	2340	PUBLISHED
0H102A	10*25*78	- 11*29*78	8/ 13	0	5/140C	(56)	AEDC B / HYPERSONIC	B65	2455	PUBLISHED
0H103A	2*20*78	- 02*21*78	12/ 8	72	VEH. 5 F'BODY	(83)	AEDC B / HYPERSONIC	V2C	2420	PUBLISHED
0H103B	4*27*78	- 04*28*78	24/ 12	53	5/140C	(60)	AEDC D / HYPERSONIC	V2C	2427	PUBLISHED
0H105A	5*15*79	- 06*20*79	24/ 24	0	5/140C	(60)	AEDC B / HYPERSONIC	B67	2464	PUBLISHED
0H105B	7*23*79	- 08*01*79	24/ 180	0	5/140C	(60)	ARC 3.5-FT HYPERSONIC	247	2468	PUBLISHED
0H107	1* 7*81	- 01*08*81	12/ 12	0	ELEV/ELEV	(94)	AEDC B / HYPERSONIC	B17	2492	PUBLISHED
0H108	12*15*80	- 01*15*81	200/ 139	43	ELEV/ELEV	(93)	ARC 3.5-FT HYPERSONIC	254	2494	PUBLISHED

TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
OH109	10*27*80 - 11*24*80	48/ 40	0	5/140C	(56)	AEDC B / HYPERSONIC	G9	2490	PUBLISHED
OH110	11*17*80 - 01*30*81	80/ 200	0	5/140C	(60)	ARC 3.5-FT HYPERSONIC	253	2495	PUBLISHED
OH111	9*24*81 - 09*30*81	32/ 32	0	5/140C	(60)	AEDC B / HYPERSONIC	1C	2496	PUBLISHED
OH400	8* 1*79 - 09*01*79	36/ 36	124	140C SILTS	(92)	ARC 11-FT TRANSONIC	B65	2472	PUBLISHED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
IH1	12* 3*73 - 12*14*73	100/ 100	44	3/139	(22)	LARC UNITARY PLAN	1071	2153	PUBLISHED
IH2	9* 4*73 - 09*11*73	80/ 104	21	3/139	(26)	ARC 3.5-FT HYPERSONIC	171	2085	PUBLISHED
IH3	10*31*73 - 11*09*73	128/ 128	79	3/139	(22)	ARC 3.5-FT HYPERSONIC	178	2136	PUBLISHED
IH4	11*12*73 - 11*16*73	40/ 64	47	3/139	(26)	LARC UNITARY PLAN	1059	2138	PUBLISHED
IH5	1*21*74 - 07*22*74	120/ 105	106	2A/089B	(19)	CALSPAN 32-IN LUDWIEG	181	2308	PUBLISHED
IH11	4* 1*78 - 04*18*78	80/ 64	0	5/140C	(84)	LERC 10X10-FT SUPERSONIC	045	2428	PUBLISHED
IH15	8*13*73 - 08*17*73	64/ 72	30	2A/089B	(41)	ARC 3.5-FT HYPERSONIC	172	2098	PUBLISHED
IH16	7* 6*73 - 07*13*73	35/ 80	12	2A/089B	(41)	LARC UNITARY PLAN	1041	2166	PUBLISHED
IH17	10* 9*73 - 10*16*73	40/ 48	59	2A/089B	(41)	LARC MACH 8 VARIABLE DENSITY	646	2105	PUBLISHED
IH18	10*19*73 - 10*30*73	40/ 40	22	2A/089B	(41)	LARC 20-IN FREON	118	2110	PUBLISHED
IH19A	12*14*73 - 12*26*73	40/ 40	22	2A/089B	(50)	LARC HYPERSONIC NITROGEN	28	2157	PUBLISHED
IH19B	12*27*73 - 01*08*74	20/ 40	22	2A/089B	(50)	LARC HYPERSONIC NITROGEN	28	2157	PUBLISHED
IH20	1*18*74 - 02*06*74	192/ 192	105	3/139	(22)	ARC 3.5-FT HYPERSONIC	185	2148	PUBLISHED
IH21	10*29*73 - 12*13*73	80/ 145	31	3/139	(37)	CALSPAN HYPERSONIC SHOCK	100	2164	PUBLISHED
IH27	9* 7*74 - 09*25*74	80/ 196	65	TPS TILES	(15)	ARC 3.5-FT HYPERSONIC	200	2210	PUBLISHED
IH28-1	5*20*74 - 05*24*74	80/ 50	24	2A/089B	(50)	ARC 3.5-FT HYPERSONIC	195	2180	PUBLISHED
IH28-2	5*20*74 - 05*24*74	30/ 38	15	2A/089B	(50)	ARC 3.5-FT HYPERSONIC	195	2180	PUBLISHED
IH33A	10*14*74 - 10*18*74	32/ 32	10	5/140C	(37)	CALSPAN HYPERSONIC SHOCK	120	2249	PUBLISHED
IH33B	12* 5*74 - 12*19*74	48/ 80	24	5/140C	(37)	CALSPAN HYPERSONIC SHOCK	131	2249	PUBLISHED
IH34	5* 5*75 - 09*03*75	240/ 264	57	5/140C	(19)	LERC 10X10-FT SUPERSONIC	038	2282	PUBLISHED
IH39	9*22*76 - 04*14*77	240/ 226	163	5/140C	(19)	LERC 10X10-FT SUPERSONIC	041	2435	PUBLISHED
IH41A	3*31*75 - 5*21*75	48/ 57	318	5/140C	(60)	AEDC A / SUPERSONIC	4A	2240	PUBLISHED
IH41B	12*11*75 - 01*09*76	78/ 80	300	5/140C	(60)	AEDC A / SUPERSONIC	4A	2295	PUBLISHED
IH42	4*26*76 - 05*26*76	192/ 218	57	5/140C	(56)	ARC 3.5-FT HYPERSONIC	217		UNASSIGNED
IH43	12*17*75 - 02*23*76	120/ 250	60	5/140C	(59)	CALSPAN HYPERSONIC SHOCK	189	2319	PUBLISHED

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
IH47	3* 8*76	- 03*19*76	40/ 31	178	5/140C	(60)	AEDC A / SUPERSONIC	J3A	2312	PUBLISHED
IH48	4*17*75	- 05*08*75	240/ 256	125	5/140C	(60)	ARC 3.5-FT HYPERSONIC	211	2248	PUBLISHED
IH51A	7*14*77	- 07*27*77	80/ 118	62	FLAT PLATE	(58)	ARC 3.5-FT HYPERSONIC	228	2393	PUBLISHED
IH51B	7*15*78	- 07*24*78	60/ 60	0	FLAT PLATE	(58)	ARC 3.5-FT HYPERSONIC	239	2429	PUBLISHED
IH51C	12*26*78	- 02*16*79	500/ 632	0	FLAT PLATE	(58)	ARC 3.5-FT HYPERSONIC	241	2448	PUBLISHED
IH51D	5* 1*79	- 06*00*79	240/ 240	0	FLAT PLATE	(58)	ARC 3.5-FT HYPERSONIC	244	2461	PUBLISHED
IH68	10*12*76	- 12*08*76	400/ 628	0	5/140C	(60)	ARC 3.5-FT HYPERSONIC	222	2357	PUBLISHED
IH72	1* 3*77	- 01*10*77	60/ 56	0	5/140C	(60)	AEDC A / SUPERSONIC	K2A	2372	PUBLISHED
IH73	12* 1*77	- 01*23*78	160/ 160	0	5/140C	(50)	ARC 3.5-FT HYPERSONIC	233	2407	PUBLISHED
IH75	10* 3*77	- 12*12*77	200/ 320	41	5/140C	(19)	CALSPAN 32-IN LUDWIEG	100	2453	PUBLISHED
IH83	1*25*78	- 03*10*78	200/ 102	41	5/140C	(19)	LERC 10X10-FT SUPERSONIC	044	2440	PUBLISHED
IH85	4*19*78	- 04*26*78	60/ 65	337	5/140C	(60)	AEDC A / SUPERSONIC	W5	2431	PUBLISHED
IH90	1*30*78	- 03*10*78	160/ 116	73	5/140C	(60)	ARC 3.5-FT HYPERSONIC	234	2412	PUBLISHED
IH99	8*28*77	- 09*07*77	80/ 79	0	5/140C	(98)	ARC 3.5-FT HYPERSONIC	230	2452	PUBLISHED
IH100	6*20*77	- 06*23*77	16/ 32	9	GAS TEMP PROBE	(0)	ARC 3.5-FT HYPERSONIC	227	2418	PUBLISHED
IH102-1	5* 1*79	- 06*01*79	26/ 26	0	5/140C	(60)	AEDC A / SUPERSONIC	B67	2464	PUBLISHED
IH102-2	5* 1*79	- 06*01*79	12/ 12	0	5/140C	(56)	AEDC A / SUPERSONIC	B67	2464	PUBLISHED
IH102-3	5* 1*79	- 06*01*79	10/ 10	0	5/140C	(83)	AEDC A / SUPERSONIC	B67	2464	PUBLISHED
IH103-1	10* 1*79	- 11*01*79	100/ 100	0	5/140C	(60)	ARC 3.5-FT HYPERSONIC	245	2467	PUBLISHED
IH103-2	10*15*79	- 11*01*79	100/ 100	0	5/140C	(56)	ARC 3.5-FT HYPERSONIC	245	2467	PUBLISHED
IH104	2* 7*80	- 04*17*80	80/ 80	0	5/140C	(60)	ARC 3.5-FT HYPERSONIC	250	2480	PUBLISHED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

AUG 01, 1984

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
SH12F	7*29*74 -	08*07*74	80/ 80	42	SRB	(0)	LARC UNITARY PLAN	1115	2216	PUBLISHED
SH13F	10*31*75 -	11*18*75	80/ 156	124	SRB	(0)	LARC UNITARY PLAN	1159		UNASSIGNED
SH15F	12*29*75 -	02*20*76	12/ 100	0	SRB	(0)	AEDC A / SUPERSONIC	EOA		UNASSIGNED
SH16F	3*10*76 -	04*19*76	12/ 8	0	SRB	(0)	AEDC A / SUPERSONIC	E6A		UNASSIGNED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	REF.	MODEL	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
TH1F	9* 1*74 - 09*09*74	80/ 40	0	ET		(0)	AEDC F / HYPERSONIC	25A	2218	PUBLISHED
TH2F	6* 1*75 - 06*05*75	80/ 32	0	ET		(0)	CALSPAN HYPERSONIC SHOCK	000		UNASSIGNED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER AUG 01,1984

TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
MH1	1*13*76 - 01*23*76	24/ 64	30	4/140B	(29)	AEDC F / HYPERSONIC	29A		UNASSIGNED
MH2	9* 3*75 - 01*23*76	16/ 11	22	4/140B	(29)	AEDC B / HYPERSONIC	D5A		UNASSIGNED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
FH1	11*15*72	- 01*01*73	160/ 80	200	TPS TILES	(15)	LARC HIGH RE'S NUMBER HELIUM	100		UNASSIGNED
FH10	1*21*74	- 01*29*74	32/ 32	9	3/139	(22)	AEDC F / HYPERSONIC	291	2197	PUBLISHED
FH13	9*22*75	- 09*25*75	24/ 40	0	ET/SPIKE	(0)	AEDC A / SUPERSONIC	E1A	2276	PUBLISHED
FH14	3*15*76	- 04*06*76	40/ 40	0	ET/SPIKE	(0)	ARC 3.5-FT HYPERSONIC	215	2313	PUBLISHED
FH15	5* 1*78	- 05*05*78	52/ 52	0	ET/SPIKE	(0)	AEDC A / SUPERSONIC	420	2422	PUBLISHED
FH16	7* 1*78	- 07*15*78	80/ 80	0	ET/SPIKE	(0)	ARC 3.5-FT HYPERSONIC	247	2423	PUBLISHED

APPENDIX A

TABLE A3 - WIND TUNNEL TESTING BY TEST NUMBER - STRUCTURAL DYNAMICS TESTS

AUG 01.1984

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
OS1	8* 6*73	08*10*73	80/ 72	39	2A/089B	(23)	LARC 26-IN TRANSONIC BLOWDOWN	545	2094	PUBLISHED
OS2	6* 4*73	06*07*73	120/ 24	18	ATP	(24)	LARC 26-IN TRANSONIC BLOWDOWN	544	2067	PUBLISHED
OS3	8*11*73	08*14*73	58/ 52	31	2A/089B	(11)	ARC 8X7-FT SUPERSONIC	705	2401	PUBLISHED
OS4A	9* 9*74	10*09*74	154/ 260	36	PANELS	(40)	ARC 2X2-FT TRANSONIC	041	2450	PUBLISHED
OS4B	11*18*75	01*15*76	140/ 96	0	PANELS	(40)	ARC 2X2-FT TRANSONIC	154	2450	PUBLISHED
OS6	9* 2*74	09*12*74	120/ 104	27	4/140B	(54)	LARC 16-FT TRANSONIC DYNAMIC	246	2365	PUBLISHED
OS7	8*12*74	08*30*74	120/ 120	30	4/140B	(55)	LARC 16-FT TRANSONIC DYNAMIC	246	2363	PUBLISHED
OS8A	7*11*74	07*18*74	60/ 156	96	HRSI TILE	(81)	ARC 11-FT TRANSONIC	705	2179	PUBLISHED
OS8B	7*19*74	07*29*74	60/ 120	39	HRSI TILE	(81)	ARC 9X7-FT SUPERSONIC	705	2179	PUBLISHED
OS12	1*11*76	01*29*76	80/ 40	42	LRSI TILE	(85)	ARC 2X2-FT TRANSONIC	116	2450	PUBLISHED
OS13	11*24*75	11*26*75	16/ 21	45	LRSI TILE	(85)	ARC 9X7-FT SUPERSONIC	166	2287	IN PROCESS
OS20	10*22*75	10*30*75	120/ 120	14	5/140C	(79)	LARC 16-FT TRANSONIC DYNAMIC	266		UNASSIGNED
OS21	5* 8*78	05*26*78	200/ 120	0	5/140C FLUTTER	(80)	LARC 16-FT TRANSONIC DYNAMIC	300		UNASSIGNED
OS22	4* 7*75	04*10*75	80/ 58	16	4/140A,B	(55)	LARC 16-FT TRANSONIC DYNAMIC	258		UNASSIGNED
OS31	11*22*77	11*30*77	84/ 56	55	LRSI TILES	(96)	ARC 11-FT TRANSONIC	145		PUBLISHED
OS32	7*15*76	07*27*76	96/ 80	89	TILE PANEL	(35)	ARC 2X2-FT TRANSONIC	167	2339	IN PROCESS
OS36	4*16*79	04*19*79	60/ 64	0	HRSI TILE	(96)	ARC 11-FT TRANSONIC	369	2458	PUBLISHED
OS37	5* 7*79	05*11*79	60/ 40	0	HRSI TILE	(81)	ARC 9X7-FT SUPERSONIC	369	2458	PUBLISHED
OS41	4*18*79	04*20*79	8/ 16	0	LRSI TILES	(96)	ARC 11-FT TRANSONIC	369	2458	PUBLISHED
OS42	7* 2*79	07*05*79	8/ 8	0	TPS TILES	(96)	ARC 11-FT TRANSONIC	380	2463	PUBLISHED
OS45	9* 3*79	09*03*79	8/ 8	0	TPS TILES	(96)	ARC 11-FT TRANSONIC	381	2470	PUBLISHED
OS46A	3* 4*80	03*05*80	8/ 8	0	TPS	(109)	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
OS46B	3* 6*80	03*08*80	24/ 24	0	TPS	(108)	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
OS46C	4*17*80	04*18*80	8/ 8	0	TPS	(109)	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
OS46D	8* 0*80	09*00*80	8/ 8	0	TPS	(108)	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
OS46E	9*15*80	10*20*80	8/ 8	0	TPS	(108)	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
OS46F	10* 1*80	10*02*80	8/ 8	0	TPS	(108)	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
OS46G	12*10*81	12*11*81	8/ 17	0	TPS	(108)	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
OS47	11* 1*79	12*20*79	160/ 160	0	TPS	(110)	AEDC 1-FT TRANSONIC			UNASSIGNED
OS49	1*28*81	02*04*81	40/ 44	0	TPS	(111)	AEDC 16-FT TRANSONIC	556	2483	PUBLISHED
OS50	4*30*81	06*01*81	40/ 113	0	TPS	(113)	ARC 11-FT TRANSONIC	425	2485	PUBLISHED
OS51A	11*17*80	01*29*81	50/ 77	0	TPS TILE	(96)	ARC 11-FT TRANSONIC	436	2487	PUBLISHED
OS51B	11*17*80	11*19*80	50/ 50	0	TPS TILE	(96)	ARC 11-FT TRANSONIC	436	2487	PUBLISHED
OS51C	1*27*81	01*29*81	27/ 27	0	TPS TILE	(96)	ARC 11-FT TRANSONIC	436	2487	PUBLISHED
OS53A	12*12*80	01*01*81	80/ 104	0	TPS	(717)	LARC 8-FT TRANSONIC PRESSURE	905	2503	PUBLISHED
OS53B	3*23*81	04*01*81	80/ 108	0	TPS	(719)	LARC 8-FT TRANSONIC PRESSURE	909	2503	PUBLISHED
OS55	2*23*81	03*02*81	80/ 64	0	TILE	(81)	ARC 9X7-FT SUPERSONIC	464	2465	PUBLISHED
OS56	8*26*81	08*27*81	8/ 8	0	TPS TILE	(108)	AEDC 16-FT TRANSONIC	608	2489	PUBLISHED
OS57	8*26*81	08*27*81	8/ 8	0	TILE	(81)	ARC 9X7-FT SUPERSONIC	508	2465	PUBLISHED
OS60	6* 9*81	06*09*81	8/ 8	0	TPS TILE	(96)	ARC 11-FT TRANSONIC	500	2506	PUBLISHED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

AUG 01,1984

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
IS1A	8* 2*73	08*08*73	60/ 60	21	2A/089B	(11)	ARC 11-FT TRANSONIC	705	2401	PUBLISHED
IS1B	7*23*73	08*01*73	60/ 64	4	2A/089B	(11)	ARC 9X7-FT SUPERSONIC	705	2401	PUBLISHED
IS1C	8* 9*73	08*11*73	12/ 24	3	2A/089B	(11)	ARC 8X7-FT SUPERSONIC	705	2401	PUBLISHED
IS2A	11* 7*75	11*14*75	144/ 120	53	5/140C	(84)	ARC 11-FT TRANSONIC	113	2284	PUBLISHED
IS2B	9*25*75	10*29*75	60/ 60	67	5/140C	(84)	ARC 9X7-FT SUPERSONIC	113	2284	PUBLISHED
IS4	10*18*73	10*24*73	120/ 58	94	2A/089B	(30)	LARC 26-IN TRANSONIC BLOWDOWN	547	2146	PUBLISHED
IS6A	10* 2*73	10*11*73	80/ 74	126	2A/089B	(13)	MSFC 14-IN TRANSONIC	582	2158	PUBLISHED
IS6B	3*20*73	05*27*73	50/ 50	70	2A/089B	(13)	MSFC 14-IN TRANSONIC	559	2158	PUBLISHED
IS10	9*18*78	10*08*78	200/ 128	0	5/140C FLUTTER	(80)	LARC 16-FT TRANSONIC DYNAMIC	308		UNASSIGNED
IS20	7*24*78	08*25*78	200/ 216	0	5/140C + TOWER	(100)	LARC 16-FT TRANSONIC DYNAMIC	306		UNASSIGNED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

AUG 01,1984

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
CS1	6* 9*75	06*13*75	80/ 95	165	140A,B/747	(8)	UNIV. OF WASH. LOW SPEED	1160		UNASSIGNED
CS2	6* 9*75	06*16*75	95/ 95	165	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1474		UNASSIGNED
CS3	9*12*75	09*15*75	40/ 80	129	140A,B/747	(8)	UNIV. OF WASH. LOW SPEED	1170	2338	PUBLISHED
CS4	9*29*75	10*02*75	40/ 64	95	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1490	2341	PUBLISHED
CS5	11* 3*75	11*05*75	24/ 33	192	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1493	234[illegible]	PUBLISHED
CS6	2* 5*76	02*11*76	60/ 58	203	140A,B/747	(8)	GENERAL DYNAMIC - LOW SPEED	691		UNASSIGNED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER

AUG 01, 1984

TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	REF.	MODEL (ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
SS13F	8* 4*76 - 08*25*76	120/ 162	0	SRB	(0)	ARC 14-FT TRANSONIC	196		UNASSIGNED
SS13FB	7*26*76 - 08*04*76	60/ 120	0	SRB	(0)	ARC 9X7-FT SUPERSONIC	114		UNASSIGNED
SS13FC	7*19*76 - 07*23*76	60/ 60	0	SRB	(0)	ARC 8X7-FT SUPERSONIC	114		UNASSIGNED
SS14F	1*27*75 - 02*13*75	104/ 104	0	SRB	(0)	MSFC 14-IN TRANSONIC	616		UNASSIGNED
SS15F	4*12*75 - 04*12*75	40/ 90	0	SRB	(0)	MSFC 14-IN TRANSONIC	623		UNASSIGNED
SS16F	9*22*75 - 10*20*75	120/ 158	0	SRB	(0)	MSFC 14-IN TRANSONIC	636		UNASSIGNED
SS17F	4* 4*75 - 04*05*75	12/ 12	0	SRB	(0)	AEDC A / SUPERSONIC	F5A		UNASSIGNED
SS18F	11*14*75 - 12*08*75	80/ 131	0	SRB	(0)	MSFC 14-IN TRANSONIC	638		UNASSIGNED
SS19F	5*27*75 - 06*27*75	180/ 192	0	SRB	(0)	MSFC 14-IN TRANSONIC	626		UNASSIGNED
SS20F	2*15*75 - 02*21*75	40/ 44	0	SRB	(0)	MSFC 14-IN TRANSONIC	614		UNASSIGNED
SS22F	1* 1*76 - 01*15*76	64/ 45	0	SRB	(0)	AEDC A / SUPERSONIC	F9A		UNASSIGNED
SS27	3*13*78 - 04*17*78	80/ 80	0	SRB	(0)	ARC 14-FT TRANSONIC	302		UNASSIGNED
SS28FB	6*12*78 - 06*18*78	40/ 40	0	SRB	(0)	ARC 9X7-FT SUPERSONIC	281		UNASSIGNED
SS28FC	11* 6*78 - 11*22*78	52/ 98	0	SRB	(0)	ARC 8X7-FT SUPERSONIC	281		UNASSIGNED
SS30F	7*20*77 - 09*03*77	160/ 240	0	SRB	(0)	MSFC 14-IN TRANSONIC	648		UNASSIGNED

PHASE C/D SSV WIND TUNNEL TESTING PER TEST NUMBER AUG 01, 1984

TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
FS2	2* 1*77 -	06*01*77	400/ 400	0	INT VEH	(0)	MSFC ACOUSTIC MODEL TEST FAC.			UNASSIGNED
FS5A	7* 1*73 -	10*01*73	120/ 120	0	INT. VEH.	(0)	MSFC LOW DENSITY FAC.			UNASSIGNED
FS5B	1* 1*74 -	02*01*74	120/ 120	0	INT. VEH.	(0)	MSFC LOW DENSITY FAC.			UNASSIGNED
FS5C	8* 1*74 -	08*18*74	120/ 120	0	INT. VEH.	(0)	MSFC LOW DENSITY FAC.			UNASSIGNED
FS8A	11* 1*73 -	11*08*73	40/ 32	0	PRR ASCENT	(0)	JPL 20-IN SUPERSONIC			UNASSIGNED
FS8B	8* 1*74 -	08*07*74	32/ 32	0	PRR ASCENT	(0)	JPL 20-IN SUPERSONIC			UNASSIGNED

APPENDIX A

TABLE A4 - WIND TUNNEL TESTING BY FACILITY - NASA COMPLEXES

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01,1984

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
ARC 40X80-FT SUBSONIC	462	OA100	5*27*75 - 06*14*75	240/272	190	OV101(ALT)	(76)	2261	PUBLISHED
ARC 40X80-FT SUBSONIC	471	AA2	10*30*75 - 11*01*75	40/ 40	0	TUNNEL CALIB.	(0)		UNASSIGNED
ARC 40X80-FT SUBSONIC	473	OA164	11*28*75 - 12*01*75	80/ 80	22	OV101(ALT)	(76)	2499	PUBLISHED
ARC 40X80-FT SUBSONIC	479	OA174	2* 2*76 - 02*27*76	240/264	165	OV101(ALT)	(76)	2302	PUBLISHED
ARC 40X80-FT SUBSONIC	500	OA237	1*24*77 - 01*31*77	60/ 60	32	ADS PROBES	(99)	2375	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01,1984

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
ARC	12-FT PRESSURE	028	MA13	4*15*74 -	05*03*74	120/176	0	GULFSTREAM 2	(0)		UNASSIGNED
ARC	12-FT PRESSURE	078	OA159	6*23*75 -	07*08*75	160/152	50	140A,B/(ALT)	(45)	2265	PUBLISHED
ARC	12-FT PRESSURE	135	LA66	10*20*75 -	10*24*75	80/ 80	26	140C/REMOTE ELE	(44)	2281	PUBLISHED
ARC	12-FT PRESSURE	180	OA173	3*15*76 -	03*26*76	160/256	48	140C(ALT)	(45)	2304	PUBLISHED
ARC	12-FT PRESSURE	218	OA101	9*13*77 -	11*11*77	160/160	373	VEH 102	(39)	2405	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01,1984

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
ARC 2X2-FT TRANSONIC	041	OS4A	9* 9*74 - 10*09*74	154/260	36	PANELS	(40)	2450	PUBLISHED
ARC 2X2-FT TRANSONIC	154	OS4B	11*18*75 - 01*15*76	140/ 96	0	PANELS	(40)	2450	PUBLISHED
ARC 2X2-FT TRANSONIC	116	OS12	1*11*76 - 01*29*76	80/ 40	42	LRSI TILE	(85)	2450	PUBLISHED
ARC 2X2-FT TRANSONIC	167	OS32	7*15*76 - 07*27*76	96/ 80	89	TILE PANEL	(35)	2339	IN PROCESS
ARC 2X2-FT TRANSONIC	382	OA252	10*16*79 - 11*14*79	360/360	0	TILE	(107)	2473	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01,1984

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
ARC 6X6-FT SUPERSONIC	650	OA3	10*24*72 - 11*10*72	200/320	214	ATP	(6)	2009	PUBLISHED
ARC 6X6-FT SUPERSONIC	706	OA43	4*18*73 - 05*C1*73	128/160	137	2A/089B	(18)	2050	PUBLISHED
ARC 6X6-FT SUPERSONIC	630	IA29	9*12*73 - 09*25*73	80/184	111	4/140A,B	(36)	2077	PUBLISHED
ARC 6X6-FT SUPERSONIC	630	OA63	9*25*73 - 9*28*73	64/ 80	98	4/140A,B	(36)	2077	PUBLISHED
ARC 6X6-FT SUPERSONIC	709	OA59	3*13*74 - 3*21*74	120/293	150	4/140A,B	(49)	2159	PUBLISHED
ARC 6X6-FT SUPERSONIC	033	FA6	7*10*74 - 08*19*74	40/584	0	OGIVE CYL	(0)		UNASSIGNED
ARC 6X6-FT SUPERSONIC	033	FA7	8*20*74 - 08*28*74	40/166	0	OGIVE CYL	(0)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01, 1984

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
ARC 14-FT TRANSONIC	711	IA8	2*12*73 - 03*12*73	80/160	54	ATP	(6)	2173	PUBLISHED
ARC 14-FT TRANSONIC	085	CA23A	3*21*75 - 04*17*75	120/213	71	140C(MOD)/747	(48)	2243	PUBLISHED
ARC 14-FT TRANSONIC	085	CA23B	5* 1*75 - 07*22*75	160/132	46	140C(MOD)/747	(48)	2275	PUBLISHED
ARC 14-FT TRANSONIC	150	OA220	11*11*75 - 11*21*75	120/110	142	VEH 101 (ADS)	(57)	2286	PUBLISHED
ARC 14-FT TRANSONIC	143	IA137	4*26*76 - 05*03*76	40/ 56	43	ET FORETANK	(68)	2316	PUBLISHED
ARC 14-FT TRANSONIC	121	CA13	6* 8*76 - 07*01*76	160/193	54	140C(ALT)/747	(45)	2332	PUBLISHED
ARC 14-FT TRANSONIC	196	SS13F	8* 4*76 - 08*25*76	120/162	0	SRB	(0)		UNASSIGNED
ARC 14-FT TRANSONIC	302	SS27	3*13*78 - 04*17*78	80/ 80	0	SRB	(0)		UNASSIGNED

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
ARC 11-FT TRANSONIC	608	AA3A	1*10*71 - 01*31*71	300/300	0	PRE-ATP	(0)	2255	PUBLISHED
ARC 11-FT TRANSONIC	686	IA7	2*12*73 - 02*23*73	80/160	85	PRE-ATP/001	(7)	2024	PUBLISHED
ARC 11-FT TRANSONIC	707	IA9A	4* 2*73 - 04*14*73	90/113	118	2A/089B	(17)	2032	PUBLISHED
ARC 11-FT TRANSONIC	707	OA12A	4*12*73 - 04*23*73	90/103	98	2A/089B	(17)	2032	PUBLISHED
ARC 11-FT TRANSONIC	705	IS1A	8* 2*73 - 08*08*73	60/ 60	21	2A/089B	(11)	2401	PUBLISHED
ARC 11-FT TRANSONIC	716	IA14A	9* 4*73 - 09*13*73	130/151	149	4/140A,B	(47)	2084	PUBLISHED
ARC 11-FT TRANSONIC	716	OA22A	9*12*73 - 09*14*73	20/ 21	24	4/140A,B	(47)	2130	PUBLISHED
ARC 11-FT TRANSONIC	747	OA53A	11*19*73 - 11*27*73	128/128	267	4/140A,B	(47)	2128	PUBLISHED
ARC 11-FT TRANSONIC	003	MA17	4* 8*74 - 04*22*74	120/ 32	0	GULFSTREAM 2	(0)		UNASSIGNED
ARC 11-FT TRANSONIC	705	OS8A	7*11*74 - 07*18*74	60/156	96	HRSI TILE	(81)	2179	PUBLISHED
ARC 11-FT TRANSONIC	019	IA81A	7*26*74 - 08*27*74	84/184	99	4/140A,B (MOD)	(47)	2169	PUBLISHED
ARC 11-FT TRANSONIC	014	IA19A	9*16*74 - 09*23*74	156/136	201	5/140C	(88)	2170	PUBLISHED
ARC 11-FT TRANSONIC	023	IA80	11* 4*74 - 11*08*74	100/144	380	5/140C	(88)	2212	PUBLISHED
ARC 11-FT TRANSONIC	094	OA161A	3*10*75 - 03*20*75	140/160	285	140A,B (MOD)	(45)	2245	PUBLISHED
ARC 11-FT TRANSONIC	073	OA148	5* 5*75 - 05*17*75	220/264	474	4/140A,B (MOD)	(47)	2254	PUBLISHED
ARC 11-FT TRANSONIC	072	IA72	5*19*75 - 05*31*75	120/200	176	5/140C	(88)	2258	PUBLISHED
ARC 11-FT TRANSONIC	113	IS2A	11* 7*75 - 11*14*75	144/120	53	5/140C	(84)	2284	PUBLISHED
ARC 11-FT TRANSONIC	144	IA135A	3* 2*76 - 03*23*76	120/146	132	4/140A,B (MOD)	(47)	2306	PUBLISHED
ARC 11-FT TRANSONIC	074	SA11FA	4*14*76 - 04*26*76	200/193	0	SRB	(483)	2331	PUBLISHED
ARC 11-FT TRANSONIC	187	OA175	6*28*76 - 07*09*76	160/240	290	140A,B (ALT)	(47)	2333	PUBLISHED
ARC 11-FT TRANSONIC	200	LA77	7* 9*76 - 07*24*76	120/151	521	140C/REMOTE ELE	(44)	2344	PUBLISHED
ARC 11-FT TRANSONIC	103	FA23C	7*24*76 - 08*04*76	60/ 44	0	TRIPLE BODY	(0)		UNASSIGNED
ARC 11-FT TRANSONIC	213	LA89	10* 4*76 - 10*18*76	160/208	220	OV101 (ALT)	(201)	2353	PUBLISHED
ARC 11-FT TRANSONIC	115	OA149A	2*24*77 - 03*04*77	160/144	390	5/140C	(47)	2376	PUBLISHED
ARC 11-FT TRANSONIC	118	OA145A	3* 8*77 - 04*02*77	160/480	981	VEH 102	(39)	2380	PUBLISHED
ARC 11-FT TRANSONIC	228	IA144	4* 6*77 - 04*15*77	160/200	514	5/140C	(72)	2377	PUBLISHED

ARC 11-FT TRANSONIC	275	IA119	10* 7*77 - 10*31*77	170/285	620	5/140C	(88)	2404	PUBLISHED
ARC 11-FT TRANSONIC	145	OS31	11*22*77 - 11*30*77	84/ 56	55	LRSI TILES	(96)		PUBLISHED
ARC 11-FT TRANSONIC	289	OA126A	5* 1*78 - 05*30*78	240/131	304	5/140C	(47)	2424	PUBLISHED
ARC 11-FT TRANSONIC	369	OS36	4*16*79 - 04*19*79	60/ 64	0	HRSI TILE	(96)	2458	PUBLISHED
ARC 11-FT TRANSONIC	369	OS41	4*18*79 - 04*20*79	8/ 16	0	LRSI TILES	(96)	2458	PUBLISHED
ARC 11-FT TRANSONIC	380	OS42	7* 2*79 - 07*05*79	8/ 8	0	TPS TILES	(96)	2463	PUBLISHED
ARC 11-FT TRANSONIC	B65	OH400	8* 1*79 - 09*01*79	36/ 36	124	140C SILTS	(92)	2472	PUBLISHED
ARC 11-FT TRANSONIC	381	OS45	9* 3*79 - 09*03*79	8/ 8	0	TPS TILES	(96)	2470	PUBLISHED
ARC 11-FT TRANSONIC	411	IA190A	2* 7*80 - 02*21*80	160/168	166	5/140C	(47)	2476	IN PROCESS
ARC 11-FT TRANSONIC	427	OA400	4*23*80 - 05*02*80	120/120	200	5/140C	(47)	2482	PUBLISHED
ARC 11-FT TRANSONIC	412	IA191	6*20*80 - 06*27*80	40/ 40	0	FUEL LINE	(112)	2378	PUBLISHED
ARC 11-FT TRANSONIC	436	OS51B	11*17*80 - 11*19*80	50/ 50	0	TPS TILE	(96)	2487	PUBLISHED
ARC 11-FT TRANSONIC	436	OS51A	11*17*80 - 01*29*81	50/ 77	0	TPS TILE	(96)	2487	PUBLISHED
ARC 11-FT TRANSONIC	436	OS51C	1*27*81 - 01*29*81	27/ 27	0	TPS TILE	(96)	2487	PUBLISHED
ARC 11-FT TRANSONIC	425	OS50	4*30*81 - 06*01*81	40/113	0	TPS	(113)	2485	PUBLISHED
ARC 11-FT TRANSONIC	500	OS60	6* 9*81 - 06*09*81	8/ 8	0	TPS TILE	(96)	2506	PUBLISHED
ARC 11-FT TRANSONIC	510	MA33A	4*19*82 - 04*30*82	80/144	0	VEH 102	(106)	2507	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01,1984

FACILITY	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
ARC 9X7-FT SUPERSONIC	608	AA3B	2* 1*71 - 02*15*71	100/100	0	PRE-ATP	(0)	2255	PUBLISHED
ARC 9X7-FT SUPERSONIC	616	IA2	10*11*72 - 11*03*72	40/244	92	PRE-ATP/001	(7)	2013	PUBLISHED
ARC 9X7-FT SUPERSONIC	710	IA12B	4*23*73 - 05*07*73	120/156	63	2A/089B(MOD)	(14)	2048	PUBLISHED
ARC 9X7-FT SUPERSONIC	707	IA9B	5* 2*73 - 05*09*73	100/120	65	2A/089B	(17)	2032	PUBLISHED
ARC 9X7-FT SUPERSONIC	705	IS1B	7*23*73 - 08*01*73	60/ 64	4	2A/089B	(11)	2401	PUBLISHED
ARC 9X7-FT SUPERSONIC	716	IA14B	9*14*73 - 09*19*73	48/ 41	66	4/140A,B	(47)	2129	PUBLISHED
ARC 9X7-FT SUPERSONIC	716	OA22B	9*19*73 - 09*20*73	40/ 31	30	4/140A,B	(47)	2131	PUBLISHED
ARC 9X7-FT SUPERSONIC	747	OA53B	11*12*73 - 11*16*73	60/160	103	4/140A,B	(47)	2178	PUBLISHED
ARC 9X7-FT SUPERSONIC	052	IA110-2	7* 8*74 - 07*11*74	30/ 20	17	4/140A,B	(67)	2189	PUBLISHED
ARC 9X7-FT SUPERSONIC	052	IA110-1	7* 8*74 - 07*11*74	50/ 60	79	4/140A,B	(49)	2189	PUBLISHED
ARC 9X7-FT SUPERSONIC	705	OS8B	7*19*74 - 07*29*74	60/120	39	HRSI TILE	(81)	2179	PUBLISHED
ARC 9X7-FT SUPERSONIC	019	IA81B	8* 9*74 - 08*22*74	60/208	88	4/140A,B (MOD)	(47)	2194	PUBLISHED
ARC 9X7-FT SUPERSONIC	044	IA82B	1*28*75 - 02*04*75	70/132	286	5/140C	(75)	2231	PUBLISHED
ARC 9X7-FT SUPERSONIC	094	OA161B	3*20*75 - 03*26*75	24/ 30	49	140A,B (MOD)	(45)	2245	PUBLISHED
ARC 9X7-FT SUPERSONIC	113	IS2B	9*25*75 - 10*29*75	60/ 60	67	5/140C	(84)	2284	PUBLISHED
ARC 9X7-FT SUPERSONIC	166	OS13	11*24*75 - 11*26*75	16/ 21	45	LRSI TILE	(85)	2287	IN PROCESS
ARC 9X7-FT SUPERSONIC	103	FA23B	12* 1*75 - 12*12*75	60/ 72	0	TRIPLE BODY	(0)		UNASSIGNED
ARC 9X7-FT SUPERSONIC	144	IA135B	3* 5*76 - 03*23*76	60/100	50	4/140A,B (MOD)	(47)	2306	PUBLISHED
ARC 9X7-FT SUPERSONIC	074	SA11FB	4*25*76 - 05*07*76	120/ 84	0	SRB	(483)	2331	PUBLISHED
ARC 9X7-FT SUPERSONIC	114	SS13FB	7*26*76 - 08*04*76	60/120	0	SRB	(0)		UNASSIGNED
ARC 9X7-FT SUPERSONIC	119	OA221B	11* 8*76 - 21*15*76	60/ 76	184	ADS PROBES	(99)	2360	PUBLISHED
ARC 9X7-FT SUPERSONIC	115	OA149B	2* 2*77 - 02*07*77	40/168	201	5/140C	(47)	2370	PUBLISHED
ARC 9X7-FT SUPERSONIC	113	OA145B	4*15*77 - 05*03*77	80/348	240	VEH 102	(39)	2364	PUBLISHED
ARC 9X7-FT SUPERSONIC	272	IA156B	12*16*77 - 01*06*78	100/191	177	VEH 102	(89)	2408	PUBLISHED
ARC 9X7-FT SUPERSONIC	242	IA105B	1* 9*78 - 02*01*78	100/258	143	5/140C	(47)	2413	PUBLISHED
ARC 9X7-FT SUPERSONIC	282	OA251B	4*17*78 - 04*23*78	40/ 80	90	ADS PROBES	(99)	2421	PUBLISHED

ARC 9X7-FT SUPERSONIC	289	OA126B	4*17*78 - 04*30*78	120/ 97	256	5/140C	(47)	2424	PUBLISHED
ARC 9X7-FT SUPERSONIC	281	SS28FB	6*12*78 - 06*18*78	40/ 40	0	SRB	(0)		UNASSIGNED
ARC 9X7-FT SUPERSONIC	246	IA138	8*21*78 - 09*01*78	70/112	224	5/140C	(75)	2438	PUBLISHED
ARC 9X7-FT SUPERSONIC	283	IA131B	11* 3*78 - 11*09*78	48/ 40	0	ET FORETANK	(68)	2462	PUBLISHED
ARC 9X7-FT SUPERSONIC	347	IA184	4* 2*79 - 04*13*79	24/ 40	115	5/140C	(47)	2456	PUBLISHED
ARC 9X7-FT SUPERSONIC	369	OS37	5* 7*79 - 05*11*79	60/ 40	0	HRSI TILE	(81)	2458	PUBLISHED
ARC 9X7-FT SUPERSONIC	411	IA190B	5*20*80 - 02*21*80	120/104	294	5/140C	(47)	2476	IN PROCESS
ARC 9X7-FT SUPERSONIC	464	OS55	2*23*81 - 03*02*81	80/ 64	0	TILE	(81)	2465	PUBLISHED
ARC 9X7-FT SUPERSONIC	508	OS57	8*26*81 - 08*27*81	8/ 8	0	TILE	(81)	2465	PUBLISHED
ARC 9X7-FT SUPERSONIC	513	MA35B	12* 2*81 - 12*16*81	40/ 80	0	ADS PROBES	(99)		UNASSIGNED
ARC 9X7-FT SUPERSONIC	510	MA33B	5*10*82 - 05*21*82	40/ 96	0	VEH 102	(106)	2507	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01,1984

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
ARC 8X7-FT SUPERSONIC	707	IA9C	4*22*73 - 05*01*73	60/ 60	102	2A/089B	(17)	2032	PUBLISHED
ARC 8X7-FT SUPERSONIC	707	OA12C	5* 2*73 - 05*10*73	60/ 60	46	2A/089B	(17)	2032	PUBLISHED
ARC 8X7-FT SUPERSONIC	710	IA12C	7*11*73 - 07*27*73	220/220	133	2A/089(MOD)	(14)	2065	PUBLISHED
ARC 8X7-FT SUPERSONIC	705	IS1C	8* 9*73 - 08*11*73	12/ 24	3	2A/089B	(11)	2401	PUBLISHED
ARC 8X7-FT SUPERSONIC	705	OS3	8*11*73 - 08*14*73	58/ 52	31	2A/089B	(11)	2401	PUBLISHED
ARC 8X7-FT SUPERSONIC	747	OA53C	11*28*73 - 12*06*73	60/159	159	4/140A,B	(47)	2185	PUBLISHED
ARC 8X7-FT SUPERSONIC	044	IA82C	11*11*74 - 11*15*74	80/ 92	240	5/140C	(75)	2219	PUBLISHED
ARC 8X7-FT SUPERSONIC	094	OA161C	3*26*75 - 03*31*75	20/ 22	45	140A,B (MOD)	(45)	2245	PUBLISHED
ARC 8X7-FT SUPERSONIC	144	IA135C	3*12*76 - 03*23*76	20/ 40	5	4/140A,B (MOD)	(47)	2306	PUBLISHED
ARC 8X7-FT SUPERSONIC	074	SA11FC	3*29*76 - 04*14*76	120/156	0	SRB	(483)	2331	PUBLISHED
ARC 8X7-FT SUPERSONIC	114	SS13FC	7*19*76 - 07*23*76	60/ 60	0	SRB	(0)		UNASSIGNED
ARC 8X7-FT SUPERSONIC	119	OA221C	11*15*76 - 11*22*76	60/ 68	58	ADS PROBES	(99)	2360	PUBLISHED
ARC 8X7-FT SUPERSONIC	115	OA149C	2*16*77 - 02*18*77	40/144	25	5/140C	(47)	2370	PUBLISHED
ARC 8X7-FT SUPERSONIC	118	OA145C	4* 6*77 - 04*20*77	80/100	188	VEH 102	(39)	2389	PUBLISHED
ARC 8X7-FT SUPERSONIC	282	OA251C	5*29*78 - 06*15*78	40/ 72	96	ADS PROBES	(99)	2421	PUBLISHED
ARC 8X7-FT SUPERSONIC	281	SS28FC	11* 6*78 - 11*22*78	52/ 98	0	SRB	(0)		UNASSIGNED
ARC 8X7-FT SUPERSONIC	318	OA146	11*28*78 - 12*07*78	80/116	30	5/140C	(47)	2445	PUBLISHED
ARC 8X7-FT SUPERSONIC	289	OA126C	12* 8*78 - 12*22*78	80/ 56	134	5/140C	(47)	2424	PUBLISHED
ARC 8X7-FT SUPERSONIC	283	IA'31C	3* 5*79 - 03*11*79	48/ 40	0	ET FORETANK	(68)	2462	PUBLISHED
ARC 8X7-FT SUPERSONIC	513	.A.35C	4*19*82 - 04*23*82	40/120	0	ADS PROBES	(99)		UNASSIGNED

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
ARC 3.5-FT HYPERSONIC	147	OA4	10* 2*72 - 10*17*72	200/176	54	ATP	(6)	2007	PUBLISHED
ARC 3.5-FT HYPERSONIC	156	MA6	4* 2*73 - 04*06*73	120/136	4	RI PRR ORB.	(27)		UNASSIGNED
ARC 3.5-FT HYPERSONIC	157	OA11A	4* 9*73 - 04*17*73	144/176	62	2A/089B	(18)	2044	PUBLISHED
ARC 3.5-FT HYPERSONIC	158	OH2	4*18*73 - 06*01*73	40/144	81	TPS TILES	(15)	2035	PUBLISHED
ARC 3.5-FT HYPERSONIC	160	OA11B	5*14*73 - 05*25*73	140/160	70	2A/089B	(18)	2059	PUBLISHED
ARC 3.5-FT HYPERSONIC	163	OA58	6* 4*73 - 06*18*73	80/ 76	38	3/139B	(42)	2060	PUBLISHED
ARC 3.5-FT HYPERSONIC	167	OA73	7*11*73 - 7*18*73	60/ 96	37	3/139B	(42)	2082	PUBLISHED
ARC 3.5-FT HYPERSONIC	168	OA23-1	7*19*73 - 07*31*73	80/ 54	23	3A/140A	(49)	2071	PUBLISHED
ARC 3.5-FT HYPERSONIC	168	OA23-2	7*26*73 - 07*31*73	80/ 90	39	3/139B	(32)	2071	PUBLISHED
ARC 3.5-FT HYPERSONIC	169	IA10	8* 1*73 - 08*03*73	50/ 40	18	3/139B	(32)	2078	PUBLISHED
ARC 3.5-FT HYPERSONIC	172	IH15	8*13*73 - 08*17*73	64/ 72	30	2A/089B	(41)	2098	PUBLISHED
ARC 3.5-FT HYPERSONIC	171	OH10	8*17*73 - 09*04*73	96/104	35	3/139	(26)	2085	PUBLISHED
ARC 3.5-FT HYPERSONIC	171	IH2	9* 4*73 - 09*11*73	80/104	21	3/139	(26)	2085	PUBLISHED
ARC 3.5-FT HYPERSONIC	173	OH15	9*12*73 - 09*20*73	64/ 96	32	FLAT PLATE	(53)	2385	PUBLISHED
ARC 3.5-FT HYPERSONIC	175	IA15	10*10*73 - 10*16*73	64/ 80	25	3/139B	(32)	2102	PUBLISHED
ARC 3.5-FT HYPERSONIC	176	OA87	10*15*73 - 10*23*73	80/ 80	30	4/140A,B	(49)	2115	PUBLISHED
ARC 3.5-FT HYPERSONIC	177	OH44	10*24*73 - 10*30*73	80/ 80	46	FLAT PLATE	(53)	2386	PUBLISHED
ARC 3.5-FT HYPERSONIC	178	IH3	10*31*73 - 11*09*73	128/128	79	3/139	(22)	2136	PUBLISHED
ARC 3.5-FT HYPERSONIC	180	IA16	11*17*73 - 12*04*73	80/ 52	9	4/140A,B	(36)	2124	PUBLISHED
ARC 3.5-FT HYPERSONIC	180	OA26	11*17*73 - 12*04*73	64/140	27	4/140A,B	(36)	2124	PUBLISHED
ARC 3.5-FT HYPERSONIC	182	OH43	12* 2*73 - 12*21*73	160/128	92	TPS TILES	(15)	2250	PUBLISHED
ARC 3.5-FT HYPERSONIC	185	IH20	1*18*74 - 02*06*74	192/192	105	3/139	(22)	2148	PUBLISHED
ARC 3.5-FT HYPERSONIC	183	OH6	2* 6*74 - 02*11*74	48/ 56	39	3/139	(22)	2151	PUBLISHED
ARC 3.5-FT HYPERSONIC	186	AA1A	2*12*74 - 02*25*74	64/ 64	28	4/140A,B	(49)		UNASSIGNED
ARC 3.5-FT HYPERSONIC	187	OA36	2*25*74 - 03*01*74	80/ 80	38	4/140A,B	(49)	2162	PUBLISHED
ARC 3.5-FT HYPERSONIC	190	OA98	3*27*74 - 04*03*74	80/128	46	4/140A/B	(49)	2167	PUBLISHED

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ARC 3.5-FT HYPERSONIC	186	AA1B	4* 4*74 - 04*06*74	64/ 64	13	4/140A,B	(49)		UNASSIGNED
ARC 3.5-FT HYPERSONIC	191	IA18	4* 9*74 - 04*12*74	60/ 64	26	3/139B	(52)	2160	PUBLISHED
ARC 3.5-FT HYPERSONIC	194	OA83	5* 8*74 - 05*16*74	80/160	34	4/140A,B	(36)	2177	PUBLISHED
ARC 3.5-FT HYPERSONIC	195	IH28-2	5*20*74 - 05*24*74	30/ 38	15	2A/089B	(50)	2180	PUBLISHED
ARC 3.5-FT HYPERSONIC	195	IH28-1	5*20*74 - 05*24*74	80/ 50	24	2A/089B	(50)	2180	PUBLISHED
ARC 3.5-FT HYPERSONIC	196	TA9F	6* 3*74 - 06*15*74	128/144	0	ET	(466)	2181	PUBLISHED
ARC 3.5-FT HYPERSONIC	198	OH38	6*21*74 - 07*18*74	160/320	91	4/140B	(61)	2171	PUBLISHED
ARC 3.5-FT HYPERSONIC	199	OH26	7*22*74 - 07*29*74	80/ 96	56	4/140B	(22)	2193	PUBLISHED
ARC 3.5-FT HYPERSONIC	200	IH27	9* 7*74 - 09*25*74	80/196	65	TPS TILES	(15)	2210	PUBLISHED
ARC 3.5-FT HYPERSONIC	211	IH48	4*17*75 - 05*08*75	240/256	125	5/140C	(60)	2248	PUBLISHED
ARC 3.5-FT HYPERSONIC	215	FH14	3*15*76 - 04*06*76	40/ 40	0	ET/SPIKE	(0)	2313	PUBLISHED
ARC 3.5-FT HYPERSONIC	216	OH53A	4* 7*76 - 04*13*76	40/ 40	39	5/140C	(83)	2317	PUBLISHED
ARC 3.5-FT HYPERSONIC	216	OH53B	4*14*76 - 04*23*76	40/ 80	23	5/140C	(82)	2317	PUBLISHED
ARC 3.5-FT HYPERSONIC	217	IH42	4*26*76 - 05*26*76	192/218	57	5/140C	(56)		UNASSIGNED
ARC 3.5-FT HYPERSONIC	222	IH68	10*12*76 - 12*08*76	400/628	0	5/140C	(60)	2357	PUBLISHED
ARC 3.5-FT HYPERSONIC	227	IH100	6*20*77 - 06*23*77	16/ 32	9	GAS TEMP PROBE	(0)	2418	PUBLISHED
ARC 3.5-FT HYPERSONIC	228	IH51A	7*14*77 - 07*27*77	80/118	62	FLAT PLATE	(58)	2393	PUBLISHED
ARC 3.5-FT HYPERSONIC	230	IH99	8*28*77 - 09*07*77	80/ 79	0	5/140C	(98)	2452	PUBLISHED
ARC 3.5-FT HYPERSONIC	233	IH73	12* 1*77 - 01*23*78	160/160	0	5/140C	(50)	2407	PUBLISHED
ARC 2.5-FT HYPERSONIC	234	IH90	1*30*78 - 03*10*78	160/116	73	5/140C	(60)	2412	PUBLISHED
ARC 3.5-FT HYPERSONIC	235	OH58	3*24*78 - 04*21*78	120/168	58	ELEV/ELEV SEAL	(93)	2417	PUBLISHED
ARC 3.5-FT HYPERSONIC	247	FH16	7* 1*78 - 07*15*78	80/ 80	0	ET/SPIKE	(0)	2423	PUBLISHED
ARC 3.5-FT HYPERSONIC	239	IH51B	7*15*78 - 07*24*78	60/ 60	0	FLAT PLATE	(58)	2429	PUBLISHED
ARC 3.5-FT HYPERSONIC	241	IH51C	12*26*78 - 02*16*79	500/632	0	FLAT PLATE	(58)	2448	PUBLISHED
ARC 3.5-FT HYPERSONIC	244	IH51D	5* 1*79 - 06*00*79	240/240	0	FLAT PLATE	(58)	2461	PUBLISHED
ARC 3.5-FT HYPERSONIC	246	OH84C	6*15*79 - 06*28*79	80/ 80	0	5/140C	(60)	2468	PUBLISHED
ARC 3.5-FT HYPERSONIC	247	OH105B	7*23*79 - 08*01*79	24/180	0	5/140C	(60)	2468	PUBLISHED
ARC 3.5-FT HYPERSONIC	245	IH103-1	10* 1*79 - 11*01*79	100/100	0	5/140C	(60)	2467	PUBLISHED
ARC 3.5-FT HYPERSONIC	245	IH103-2	10*15*79 - 11*01*79	100/100	0	5/140C	(56)	2467	PUBLISHED
ARC 3.5-FT HYPERSONIC	250	IH104	2* 7*80 - 04*17*80	80/ 80	0	5/140C	(60)	2480	PUBLISHED
ARC 3.5-FT HYPERSONIC	253	OH110	11*17*80 - 01*30*81	80/200	0	5/140C	(60)	2495	PUBLISHED

ARC 3.5-FT HYPERSONIC	254	OH108	12*15*80 - 01*15*81	200/139	43	ELEV/ELEV	(93)	2494	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
LARC	7X10-FT HIGH SPEED	999	LA80	10* 6*75 -	11*07*75	156/156	83	140C/747	(88)	2299	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01, 1984

FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
LARC V/STOL	114	OA155	2*10*75 -	03*07*75	80/152	205	4/140A,B (MOD)	(47)	2237	IN PROCESS
LARC V/STOL	129	CA8	8*18*75 -	09*12*75	200/324	536	4/140A,B/747	(43)	2290	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LARC LOW TURBULANCE PRESSURE	130	LA9A	4*26*73 - 05*07*73	160/ 96	65	089B,139 NOSE	(0)	2056	PUBLISHED
LARC LOW TURBULANCE PRESSURE	135	LA9B	5*23*73 - 05*31*73	140/ 32	22	089B,139 NOSE	(0)	2056	PUBLISHED
LARC LOW TURBULANCE PRESSURE	138	OA17-1	5*18*73 - 07*06*73	60/124	65	3/139B	(42)	2058	PUBLISHED
LARC LOW TURBULANCE PRESSURE	138	OA17-2	6*18*73 - 07*06*73	20/100	55	2A/089B	(18)	2058	PUBLISHED
LARC LOW TURBULANCE PRESSURE	141	LA23	7*31*73 - 08*03*73	48/ 32	15	L/O-100 ORB.	(0)	2070	PUBLISHED
LARC LOW TURBULANCE PRESSURE	148	LA9C	10*31*73 - 11*02*73	140/ 32	28	089B,139 NOSE	(0)	2056	PUBLISHED
LARC LOW TURBULANCE PRESSURE	202	LA21A	8*19*74 - 08*30*74	160/144	55	089B, 139 NOSE	(0)		UNASSIGNED
LARC LOW TURBULANCE PRESSURE	206	LA21B	1*29*75 - 02*05*75	80/ 88	37	089B, 139 NOSE	(0)		UNASSIGNED
LARC LOW TURBULANCE PRESSURE	214	LA36B	6* 3*75 - 06*05*75	75/ 27	41	140A,B	(32)	2292	PUBLISHED
LARC LOW TURBULANCE PRESSURE	219	LA61A	8*25*75 - 09*10*75	40/ 40	138	140C/REMOTE ELE	(44)	2278	CANCEL
LARC LOW TURBULANCE PRESSURE	227	LA73A	12*18*75 - 12*30*75	82/ 82	14	4/140A,B	(69)	2298	PUBLISHED
LARC LOW TURBULANCE PRESSURE	228	LA61B	1* 5*76 - 01*14*76	40/ 96	81	140C/REMOTE ELE	(44)	2300	PUBLISHED
LARC LOW TURBULANCE PRESSURE	229	LA81	1*14*76 - 01*23*76	40/120	54	ORB/TC (ALT)	(0)	2296	PUBLISHED
LARC LOW TURBULANCE PRESSURE	238	LA73B	12*10*76 - 12*13*75	16/ 16	6	4/140A,B	(69)	2298	PUBLISHED
LARC LOW TURBULANCE PRESSURE	239	LA100A	12*15*76 - 12*17*76	14/ 14	4	GENERIC	(0)		UNASSIGNED
LARC LOW TURBULANCE PRESSURE	240	LA98	1*28*77 - 02*02*77	32/ 64	42	L.E. VORTEX	(0)		UNASSIGNED
LARC LOW TURBULANCE PRESSURE	241	LA100B	2* 2*77 - 02*07*77	25/ 25	24	GENERIC	(0)		UNASSIGNED

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LARC	8-FT TRANSONIC PRESSURE	626	LA1	11*19*72 - 12*19*72	60/ 84	73	ATP	(6)	2002	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	643	LA6	4*12*73 - 04*18*73	72/ 72	108	089B, 139 NOSE	(0)	2040	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	648	LA17	5* 4*73 - 05*14*73	80/ 96	102	L/O-100 ORB	(0)	2046	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	653	LA20A	6*22*73 - 07*06*73	160/160	81	089B,139NOSE	(202)	2107	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	658	LA20C	8*29*73 - 08*31*73	160/ 44	50	089B,139NOSE	(202)	2107	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	661	OA25	9*14*73 - 09*21*73	80/ 88	156	4/140A,B	(49)	2089	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	662	SA2FB	9*24*73 - 09*28*73	60/ 52	60	SRB	(454)	2088	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	667	IA41	12*11*73 - 12*14*73	80/ 64	86	4/140A,B	(67)	2118	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	669	LA38A	12*14*73 - 12*21*73	56/ 56	59	140A,B	(0)	2121	CANCEL
LARC	8-FT TRANSONIC PRESSURE	668	OA106	12*17*73 - 12*18*73	20/ 24	18	4/140A,B	(67)	2120	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	676	LA38B	3*27*74 - 03*29*74	48/ 4[illegible]	37	140A,B	(0)	2239	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	677	LA44	4* 2*74 - 04*09*74	160/ 96	54	4/140A,B	(0)	2200	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	680	LA48	4*10*74 - 04*15*74	48/ 48	99	089B-MOD NOSE	(0)	2184	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	684	LA51	5*24*74 - 05*31*74	80/ 72	140	140A,B	(0)	2183	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	686	OA116	6*10*74 - 06*14*74	80/ 80	81	4/140A,B	(49)	2186	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	687	OA102	6*17*74 - 06*18*74	18/ 18	10	4/140A,B	(36)	2229	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	692	LA20B	8*21*74 - 08*26*74	160/ 54	20	089B,139NOSE	(202)	2107	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	693	IA43	8*26*74 - 09*03*74	80/ 80	105	4/140A,B	(72)	2204	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	699	LA56	11*11*74 - 11*22*74	160/176	147	VEH. 5	(0)	2224	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	703	LA59	12*20*74 - 01*07*75	96/ 96	146	4/140A,B	(72)	2233	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	714	LA69	4*24*75 - 04*29*75	64/ 64	98	5/140C	(72)	2257	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	717	LA62	5*14*75 - 05*23*75	40/ 80	301	140C/REMOTE ELE	(44)	2264	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	740	LA72	3*26*76 - 03*31*76	72/ 72	30	4/140A,B	(69)	2309	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	749	IA93	5*10*76 - 05*14*76	80/ 96	255	5/140C	(72)	2326	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	758	LA91	9* 3*76 - 09*15*76	80/104	214	140C/REMOTE ELE	(44)	2352	PUBLISHED
LARC	8-FT TRANSONIC PRESSURE	764	LA92	11*11*76 - 11*19*76	80/152	67	OV101	(201)	2362	IN PROCESS

LARC 8-FT TRANSONIC PRESSURE	769	LA99	2*17*77 - 02*28*77	104/104	147	TAILCONE	(201)	2373	PUBLISHED
LARC 8-FT TRANSONIC PRESSURE	779	IA244	5*24*77 - 06*01*77	80/ 76	154	5/140C	(72)	2391	PUBLISHED
LARC 8-FT TRANSONIC PRESSURE	786	LA111	8* 3*77 - 08*05*77	95/ 40	95	140C SILTS	(44)	2395	PUBLISHED
LARC 8-FT TRANSONIC PRESSURE	787	LA113	8* 5*77 - 09*08*77	32/ 28	17	5/140C	(72)	2397	PUBLISHED
LARC 8-FT TRANSONIC PRESSURE	803	LA115	2* 1*78 - 02*06*78	45/ 45	175	140C/REMOTE ELE	(44)	2409	PUBLISHED
LARC 8-FT TRANSONIC PRESSURE	804	LA116	2* 6*78 - 02*06*78	32/ 32	0	140C	(201)	2411	CANCEL
LARC 8-FT TRANSONIC PRESSURE	865	LA143	12*21*79 - 01*08*80	80/ 88	0	VEH 102	(106)		UNASSIGNED
LARC 8-FT TRANSONIC PRESSURE	905	OS53A	12*12*80 - 01*01*81	80/104	0	TPS	(717)	2503	PUBLISHED
LARC 8-FT TRANSONIC PRESSURE	909	OS53B	3*23*81 - 04*01*81	80/108	0	TPS	(719)	2503	PUBLISHED

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PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LARC 16-FT TRANSONIC DYNAMIC	210	FA1	10*10*72 - 11*15*72	416/400	200	PRE-ATP/001	(10)		UNASSIGNED
LARC 16-FT TRANSONIC DYNAMIC	246	OS7	8*12*74 - 08*30*74	120/120	30	4/140B	(55)	2363	PUBLISHED
LARC 16-FT TRANSONIC DYNAMIC	246	OS6	9* 2*74 - 09*12*74	120/104	27	4/140B	(54)	2365	PUBLISHED
LARC 16-FT TRANSONIC DYNAMIC	258	OS22	4* 7*75 - 04*10*75	80/ 58	16	4/140A,B	(55)		UNASSIGNED
LARC 16-FT TRANSONIC DYNAMIC	266	OS20	10*22*75 - 10*30*75	120/120	14	5/140C	(79)		UNASSIGNED
LARC 16-FT TRANSONIC DYNAMIC	275	SA32F	3*22*76 - 04*02*76	150/150	94	SRB	(0)		UNASSIGNED
LARC 16-FT TRANSONIC DYNAMIC	300	OS21	5* 8*78 - 05*26*78	200/120	0	5/140C FLUTTER	(80)		UNASSIGNED
LARC 16-FT TRANSONIC DYNAMIC	306	IS20	7*24*78 - 08*25*78	200/216	0	5/140C + TOWER	(100)		UNASSIGNED
LARC 16-FT TRANSONIC DYNAMIC	308	IS10	9*18*78 - 10*08*78	200/128	0	5/140C FLUTTER	(80)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
LARC UNITARY PLAN	1002	MA5	9*15*72 - 09*25*72	80/ 60	30	PRE-ATP/001	(10)	2001	PUBLISHED
LARC UNITARY PLAN	995	LA4B	10*25*72 - 11*01*72	80/ 60	32	L/O-100 ORB.	(0)	2033	PUBLISHED
LARC UNITARY PLAN	1014	LA4A	11* 2*72 - 12*06*72	80/ 75	37	L/O-100 ORB.	(0)	2033	PUBLISHED
LARC UNITARY PLAN	1007	OA7	11*27*72 - 12*08*72	100/100	110	ATP	(6)	2014	PUBLISHED
LARC UNITARY PLAN	995	LA4C	2*19*73 - 02*23*73	80/ 50	43	L/O-100 ORB.	(0)	2033	PUBLISHED
LARC UNITARY PLAN	1023	LA8A	4*18*73 - 04*24*73	50/ 45	58	089B, 139 NOSE	(0)	2054	PUBLISHED
LARC UNITARY PLAN	1034	LA8B	5* 7*73 - 05*15*73	50/ 70	50	089B, 139 NOSE	(0)	2054	PUBLISHED
LARC UNITARY PLAN	1031	MA7	5*14*73 - 05*18*73	50/ 50	81	2A/089B	(6)	2069	PUBLISHED
LARC UNITARY PLAN	1035	OA44-1	6* 1*73 - 06*08*73	40/ 54	47	2A/089B	(18)	2057	PUBLISHED
LARC UNITARY PLAN	1035	OA44-2	6*11*73 - 06*15*73	40/ 54	36	3/139B	(42)	2057	PUBLISHED
LARC UNITARY PLAN	1039	LA8C	7* 3*73 - 07*06*73	50/ 30	14	089B, 139 NOSE	(0)	2054	PUBLISHED
LARC UNITARY PLAN	1041	IH16	7* 6*73 - 07*13*73	35/ 80	12	2A/089B	(41)	2166	PUBLISHED
LARC UNITARY PLAN	1040	LA8D	7*10*73 - 07*13*73	50/ 42	37	089B, 139 NOSE	(0)	2090	PUBLISHED
LARC UNITARY PLAN	1043	OA70	7*20*73 - 7*26*73	30/ 40	66	3/139B	(42)	2073	PUBLISHED
LARC UNITARY PLAN	1049	LA14B	8* 6*73 - 08*16*73	100/ 90	47	089B, 139 NOSE	(202)	2106	PUBLISHED
LARC UNITARY PLAN	1046	LA14A	8*17*73 - 08*28*73	100/ 80	20	089B, 139 NOSE	(202)	2106	PUBLISHED
LARC UNITARY PLAN	1055	LA14C	9* 5*73 - 09*10*73	100/ 40	45	089B, 139 NOSE	(202)	2106	PUBLISHED
LARC UNITARY PLAN	1057	OA20A	9*10*73 - 09*13*73	50/ 40	29	4/140A,B	(49)	2083	PUBLISHED
LARC UNITARY PLAN	1063	OA64	10*30*73 - 10*31*73	50/ 30	28	4/140A,B	(36)	2108	PUBLISHED
LARC UNITARY PLAN	1063	IA35	11* 1*73 - 11*02*73	60/ 30	22	4/140A,B	(36)	2108	PUBLISHED
LARC UNITARY PLAN	1057	OA20C	11* 5*73 - 11*08*73	40/ 35	19	4/140A,B	(49)	2147	PUBLISHED
LARC UNITARY PLAN	1065	LA24A	11* 9*73 - 11*12*73	40/ 20	6	089B, 139 NOSE	(202)		UNASSIGNED
LARC UNITARY PLAN	1059	IH4	11*12*73 - 11*16*73	40/ 64	47	3/139	(26)	2138	PUBLISHED
LARC UNITARY PLAN	1056	IA42A	11*27*73 - 12*04*73	40/ 70	62	4/140A,B	(67)	2119	PUBLISHED
LARC UNITARY PLAN	1071	IH1	12* 3*73 - 12*14*73	100/100	44	3/139	(22)	2153	PUBLISHED
LARC UNITARY PLAN	1058	LA14D	12* 5*73 - 12*07*73	100/ 30	9	089B, 139 NOSE	(202)	2106	PUBLISHED

LARC UNITARY PLAN	1073	IA42B	12*17*73 - 12*21*73	60/ 50	42	4/140A,B	(67)	2119	PUBLISHED
LARC UNITARY PLAN	1075	LA39A	12*26*73 - 12*28*73	50/ 24	9	140A,B	(0)	2188	PUBLISHED
LARC UNITARY PLAN	1065	LA24B	1* 2*74 - 01*07*74	40/ 34	20	089B, 139 NOSE	(202)		UNASSIGNED
LARC UNITARY PLAN	1075	LA39B	2*11*74 - 02*15*74	50/ 50	36	140A,B	(0)	2188	PUBLISHED
LARC UNITARY PLAN	1074	LA43A	3* 4*74 - 03*22*74	50/ 90	42	4/140A,B	(0)	2199	PUBLISHED
LARC UNITARY PLAN	1087	SA25F	3* 4*74 - 03*11*74	40/ 30	16	SRB	(454)	2150	PUBLISHED
LARC UNITARY PLAN	1093	LA43B	3*18*74 - 03*27*74	50/ 70	28	4/140A,B	(0)	2199	PUBLISHED
LARC UNITARY PLAN	1075	LA39C	4* 1*74 - 04*08*74	50/ 80	26	140A,B	(0)	2188	PUBLISHED
LARC UNITARY PLAN	1097	OA20B	4* 8*74 - 04*12*74	50/ 43	30	4/140A,B	(49)	2163	PUBLISHED
LARC UNITARY PLAN	1101	LA49A	4*24*74 - 04*26*74	20/ 30	37	089B-MOD NOSE	(0)	2182	PUBLISHED
LARC UNITARY PLAN	1111	LA49B	7*15*74 - 07*17*74	20/ 25	105	089B-MOD NOSE	(0)	2182	PUBLISHED
LARC UNITARY PLAN	1115	SH12F	7*29*74 - 08*07*74	80/ 80	42	SRB	(0)	2216	PUBLISHED
LARC UNITARY PLAN	1088	IA44A	8*12*74 - 08*16*74	40/ 50	27	4/140A,B	(72)	2206	PUBLISHED
LARC UNITARY PLAN	1119	IA44B	8*19*74 - 08*23*74	40/ 80	47	4/140A,B	(72)	2206	PUBLISHED
LARC UNITARY PLAN	1092	LA46A	9*13*74 - 09*24*74	96/ 96	61	140A,B ORB	(0)	2228	PUBLISHED
LARC UNITARY PLAN	1117	LA46B	9*24*74 - 10*10*74	88/ 88	51	140A,B ORB	(0)	2228	PUBLISHED
LARC UNITARY PLAN	1118	LA63A	7*18*75 - 07*18*75	40/ 14	63	140C/REMOTE ELE	(44)	2270	PUBLISHED
LARC UNITARY PLAN	1147	LA71B	7*21*75 - 07*31*75	48/ 48	30	4/140A,B	(69)	2271	PUBLISHED
LARC UNITARY PLAN	1151	LA63B	9*12*75 - 09*17*75	40/ 38	19	140C/REMOTE ELE	(34)	2279	PUBLISHED
LARC UNITARY PLAN	1132	LA71A	10*17*75 - 10*22*75	48/ 64	15	4/140A,B	(69)	2271	PUBLISHED
LARC UNITARY PLAN	1159	SH13F	10*31*75 - 11*18*75	80/156	124	SRB	(0)		UNASSIGNED
LARC UNITARY PLAN	1173	LA75	4* 6*76 - 04*16*76	90/ 90	283	140C/REMOTE ELE	(44)	2318	PUBLISHED
LARC UNITARY PLAN	1152	IA94A	4*18*76 - 04*23*76	40/ 60	92	5/140C	(72)	2323	PUBLISHED
LARC UNITARY PLAN	1177	IA94B	4*26*76 - 05*04*76	80/ 84	144	5/140C	(72)	2324	PUBLISHED
LARC UNITARY PLAN	1194	LA101	[illegible]*18*77 - 05*24*77	55/ 55	200	140C/REMOTE ELE	(44)	2390	PUBLISHED
LARC UNITARY PLAN	1207	LA124	6* 7*77 - 06*10*77	40/ 40	19	5/140C	(74)	2426	PUBLISHED
LARC UNITARY PLAN	1212	LA110	8* 8*77 - 08*10*77	30/ 30	60	140C SILTS	(44)	2396	PUBLISHED
LARC UNITARY PLAN	1217	LA114	8*23*77 - 08*31*77	30/ 60	70	140C SILTS	(44)	2399	PUBLISHED
LARC UNITARY PLAN	1243	LA125	7* 3*78 - 07*05*78	16/ 48	41	VEH. 102	(105)	2432	PUBLISHED
LARC UNITARY PLAN	1267	IA180	3*26*79 - 03*30*79	48/ 53	37	ET FORETANK	(68)	2457	PUBLISHED
LARC UNITARY PLAN	1299	LA131	1* 8*80 - 02*01*80	80/144	624	VEH 102	(106)	2478	PUBLISHED

LARC UNITARY PLAN	1311	OA255A	10*13*80 - 11*07*80	240/228	268	OV102	(70)	2498	PUBLISHED
LARC UNITARY PLAN	1358	OA255B	11* 8*80 - 11*21*80	240/132	100	OV102	(70)	2498	PUBLISHED
LARC UNITARY PLAN	1315	OA255C	11*24*80 - 12*15*80	240/140	27	OV102	(70)	2498	PUBLISHED
LARC UNITARY PLAN	1319	OA255D	1*12*81 - 02*02*81	240/160	90	OV102	(70)	2498	PUBLISHED
LARC UNITARY PLAN	1345	LA145B	9*11*81 - 09*17*81	80/ 50	37	140C	(203)	2336	PUBLISHED
LARC UNITARY PLAN	1390	LA145A	9*28*81 - 10*08*81	80/ 90	32	140C	(203)	2336	PUBLISHED
LARC UNITARY PLAN	1394	MA37	11* 2*81 - 11*04*81	40/ 24	100	VEH 102	(106)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LARC 16-FT TRANSONIC	149	LA36A	11* 5*73 - 11*11*73	75/ 75	22	140A,B	(42)		UNASSIGNED
LARC 16-FT TRANSONIC	243	SA9F	7* 8*74 - 07*29*74	150/256	90	SRB/DROGUE	(0)		UNASSIGNED
LARC 16-FT TRANSONIC	295	MA19	8*16*74 - 09*12*74	120/144	36	GULFSTREAM 2	(0)		UNASSIGNED
LARC 16-FT TRANSONIC	312	OA224	2*23*76 - 03*24*76	80/304	25	VEH 102 (ADS)	(57)	2329	PUBLISHED
LARC 16-FT TRANSONIC	325	OA270C	4* 8*78 - 04*28*78	20/ 72	80	VEH 102	(104)	2419	PUBLISHED
LARC 16-FT TRANSONIC	325	OA270B	5* 1*78 - 05*12*78	40/ 80	357	VEH. 102	(105)	2419	PUBLISHED
LARC 16-FT TRANSONIC	325	OA270A	5*15*78 - 06*09*78	60/160	156	VEH 102	(39)	2430	PUBLISHED
LARC 16-FT TRANSONIC	341	LA132	10*11*79 - 11*01*79	40/ 80	18	VEH 102	(89)	2471	PUBLISHED
LARC 16-FT TRANSONIC	342	LA140	12*26*79 - 01*03*80	80/ 80	17	VEH. 102	(105)	2475	PUBLISHED
LARC 16-FT TRANSONIC	352	OA256	2* 2*81 - 02*09*81	80/ 32	0	OV102	(70)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
LARC	MACH 8 VARIABLE DENSITY	624	LA16	6*2[illegible]*72	08*23*72	60/ 64	72	HRSI TILE	(0)	2043	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	3234	OH1A-4	9*19*72	09*26*72	10/100	120	PRE-ATP/001	(38)		UNASSIGNED
LARC	MACH 8 VARIABLE DENSITY	3234	OH1A-3	9*19*72	09*26*72	10/100	120	PRE-ATP/001	(5)		UNASSIGNED
ARC	MACH 8 VARIABLE DENSITY	3234	OH1A-2	9*19*72	09*26*72	10/100	120	PRE-ATP/001	(4)		UNASSIGNED
LARC	MACH 8 VARIABLE DENSITY	3234	OH1A-1	9*19*72	09*26*72	10/ 10	130	PRE-ATP/001	(3)		UNASSIGNED
LARC	MACH 8 VARIABLE DENSITY	3283	OH1B	11* 6*72	11*08*72	40/ 24	35	PRE-ATP/001	(4)		UNASSIGNED
LARC	MACH 8 VARIABLE DENSITY	3619	OH40	1*30*73	02*05*73	36/ 40	52	2A/089B	(31)	2049	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	3778	OH41A	3*1[illegible]*73	03*28*73	40/ 64	78	2A/089B	(33)	2075	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	4060	OH41B	5* 8*73	05*10*73	40/ 24	20	2A/089B	(38)	2076	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	4080	OH42A	5*14*73	05*16*73	20/ 20	20	3/139,139A	(46)	2101	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	4080	OH42B	5*25*73	06*01*73	40/ 48	64	3/139, 139A	(46)	2101	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	644	OH1?	6*13*73	06*13*73	8/ 8	18	2A/089B	(41)	2096	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	4080	OH42C	6*14*73	06*15*73	20/ 16	26	3/139A, W/CAN	(46)	2101	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	646	IH17	10* 9*73	10*16*73	40/ 48	59	2A/089B	(41)	2105	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	648	OH14	10*17*73	10*18*73	16/ 16	29	3A/139B	(50)	2117	PUBLISHED
LARC	MACH 8 VARIABLE DENSITY	4556	OH46	11*12*73	12*07*73	40/ 72	100	4/140B	(90)	2350	PUBLISHED

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LARC 31-IN CONT-FLOW HYP.	085	LA3	8*23*72 - 11*16*72	40/ 46	19	L/O-100 ORB.	(0)	2031	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	089	MA4	10* 1*72 - 10*02*72	16/ 16	12	RI ATP ORBITER	(0)	2008	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	096	LA11	7*11*73 - 07*20*73	24/ 58	85	089B, 139 NOSE	(0)	2066	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	097	LA32A	7*25*73 - 08*03*73	180/ 64	16	F.S. TILE ARRAY	(0)	2168	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	098	LA31	8* 9*73 - 08*16*73	48/ 72	28	LARC ORB	(0)	2047	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	099	LA13A	8*17*73 - 08*28*73	40/ 64	15	089B, 139 NOSE	(0)	2135	CANCEL
LARC 31-IN CONT-FLOW HYP.	100	LA25	8*30*73 - 09*07*73	40/ 48	126	3/139B	(32)	2126	CANCEL
LARC 31-IN CONT-FLOW HYP.	099	LA13B	10* 9*73 - 10*15*73	40/ 40	16	089B, 139 NOSE	(0)	2135	CANCEL
LARC 31-IN CONT-FLOW HYP.	101	OA85	10*31*73 - 11*08*73	50/ 60	75	3/139B	(32)	2113	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	102	LA35	11*12*73 - 11*13*73	16/ 20	19	3/139B	(32)	2127	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	099	LA13C	11*14*73 - 11*16*73	40/ 24	31	089B, 139 NOSE	(0)	2135	CANCEL
LARC 31-IN CONT-FLOW HYP.	103	LA33	11*19*73 - 11*26*73	40/ 48	26	089B, 139 NOSE	(0)		UNASSIGNED
LARC 31-IN CONT-FLOW HYP.	097	LA32B	11*28*73 - 12*03*73	180/120	43	F.S. TILE ARRAY	(0)	2168	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	104	LA47A	1* 2*74 - 01*09*74	40/120	43	140A/B ORB	(0)	2191	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	105	LA34	1*17*74 - 01*31*74	40/112	55	F.S. TILE ARRAY	(0)	2328	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	107	IA58	2* 1*74 - 02*13*74	32/ 40	34	3/139,089B	(32)	2133	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	108	IA60	2*14*74 - 02*20*74	15/ 36	55	3/139,089B	(32)	2137	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	109	OA105	2*20*74 - 2*22*74	16/ 20	50	4/140A,B	(32)	2137	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	110	OA90	3* 4*74 - 03*06*74	25/ 40	43	4/140A,B	(72)	2149	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	104	LA47B	6*10*74 - 06*24*74	40/ 88	35	140A/B ORB	(0)	2191	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	112	OH51-3	6*26*74 - 07*03*74	12/100	100	4/140B	(90)	2368	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	112	OH51-2	6*26*74 - 07*03*74	12/180	280	3/139B	(46)	2368	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	112	OH51-1	6*26*74 - 07*03*74	24/ 30	50	3/139B	(64)	2368	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	104	LA47C	7* 8*74 - 07*10*74	40/ 16	18	140A/B ORB	(0)	2191	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	113	OA82	8*12*74 - 08*16*74	40/ 48	96	4/140A,B	(32)	2195	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	114	LA57A	10* 2*74 - 10*24*74	84/144	58	140A,B	(0)	2454	PUBLISHED

LARC 31-IN CONT-FLOW HYP.	118	MA22	5* 6*75 - 06*03*75	100/168	357	4/140A,B	(32)	2267	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	114	LA57B	6* 4*75 - 06*06*75	84/ 24	10	140A,B	(0)	2454	PUBLISHED
LARC 31-IN CONT-FLOW HYP.	130	LA93	3*16*77 - 04*21*77	80/ 56	34	NOSE CONE	(0)	2383	IN PROCESS

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	REF.	MODEL (ID)	DATAMAN NO.	DATAMAN STATUS
LARC 4-FT HYPERSONIC	446	LA87	8•26•75 - 08•29•75	36/ 36	4	089B	(13)	2311	PUBLISHED
LARC 4-FT HYPERSONIC	267	LA78	1•15•76 - 01•28•76	16/ 16	4	089B	(13)	2311	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LARC 20-IN HYPERSONIC (M=6)	6441	LA15	8* 3*73 - 09*24*73	120/240	69	089B, 139 NOSE	(0)	2079	PUBLISHED
LARC 20-IN HYPERSONIC (M=6)	6456	LA54	8*14*74 - 08*19*74	28/ 28	5	140C ORB.	(0)	2213	IN PROCESS
LARC 20-IN HYPERSONIC (M=6)	6458	LA52	8*26*74 - 08*30*74	72/ 40	38	140A,B	(0)	2220	PUBLISHED
LARC 20-IN HYPERSONIC (M=6)	6468	LA88	5*21*75 - 05*21*75	16/ 16	6	089B	(13)	2311	PUBLISHED
LARC 20-IN HYPERSONIC (M=6)	6502	LA112	2* 3*77 - 02*05*77	24/ 24	0	5/140C	(0)		UNASSIGNED
LARC 20-IN HYPERSONIC (M=6)	6546	LA141A	1*12*80 - 02*01*80	80/148	0	VEH 102	(74)	2477	PUBLISHED
LARC 20-IN HYPERSONIC (M=6)	6546	LA141B	3*18*80 - 05*01*80	80/200	0	VEH 102	(74)	2477	PUBLISHED
LARC 20-IN HYPERSONIC (M=6)	6550	LA141C	6*22*80 - 07*01*80	80/ 10	0	VEH 102	(72)		UNASSIGNED
LARC 20-IN HYPERSONIC (M=6)	6559	OA257	3*12*81 - 04*20*81	80/324	380	VEH 102	(72)	2466	PUBLISHED

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LARC HYPERSONIC NITROGEN	28	IH19A	12•14•73 - 12•26•73	40/ 40	22	2A/089B	(50)	2157	PUBLISHED
LARC HYPERSONIC NITROGEN	28	IH19B	12•27•73 - 01•08•74	20/ 40	22	2A/089B	(50)	2157	PUBLISHED
LARC HYPERSONIC NITROGEN	30	OA89	7•15•74 - 08•05•74	60/143	32	5/140C	(74)	2214	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LARC 20-IN FREON	118	IH18	10*19*73 - 10*30*73	40/ 40	22	2A/089B	(41)	2110	PUBLISHED
LARC 20-IN FREON	121	OH45	11* 2*73 - 11*09*73	40/ 46	22	3A/139B	(50)	2109	PUBLISHED
LARC 20-IN FREON	220	LA53A	8*12*74 - 08*14*74	80/ 32	3	5/140C	(0)	2213	IN PROCESS
LARC 20-IN FREON	330	LA95	6* 6*77 - 10*19*77	160/160	14	NOSE CONE	(0)		UNASSIGNED
LARC 20-IN FREON	390	LA142	2* 1*80 - 03*01*80	80/ 80	0	VEH 102	(74)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LARC 22-IN HELIUM	405	LA22	6*19*72	06*30*72	160/152	31	JSC 049	(0)	2034	PUBLISHED
LARC 22-IN HELIUM	409	MA2	9*18*72	11*06*72	40/ 80	31	ATP	(1)	2003	PUBLISHED
LARC 22-IN HELIUM	411	LA2	10* 6*72	12*07*72	120/136	24	L/D-100 ORB.	(0)	2023	PUBLISHED
LARC 22-IN HELIUM	415	OA72	7*30*73	08*24*73	40/176	42	3A/139B	(34)	2092	PUBLISHED
LARC 22-IN HELIUM	418	LA12A	9* 4*73	09*17*73	40/ 80	15	089B, 139 NOSE	(0)		UNASSIGNED
LARC 22-IN HELIUM	419	LA12B	9*18*73	01*17*74	272/272	56	089B,139 NOSE	(0)		UNASSIGNED
LARC 22-IN HELIUM	7422	OA88	12*11*73	12*28*73	60/ 60	191	4/140A,B	(34)	2125	PUBLISHED
LARC 22-IN HELIUM	7426	LA40	5*13*74	06*07*74	40/ 40	25	139B	(0)	2176	PUBLISHED
LARC 22-IN HELIUM	431	OA109	8*26*74	08*29*74	60/ 88	32	5/140C	(74)	2205	PUBLISHED
LARC 22-IN HELIUM	439	LA68	2*26*75	03*20*75	120/120	26	140C ORB	(0)	2256	IN PROCESS
LARC 22-IN HELIUM	445	LA85	4* 7*76	05*24*76	88/ 88	64	140C	(13)	2343	PUBLISHED
LARC 22-IN HELIUM	306	LA53B	1*12*77	01*18*77	80/ 72	16	5/140C	(0)	2213	IN PROCESS
LARC 22-IN HELIUM	463	LA102	12* 7*78	01*04*79	40/12(	29	STING EFFECT	(0)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
LARC	26-IN TRANSONIC BLOWDOWN	544	OS2	6* 4*73 -	06*07*73	120/ 24	18	ATP	(24)	2067	PUBLISHED
LARC	26-IN TRANSONIC BLOWDOWN	545	OS1	8* 6*73 -	08*10*73	80/ 72	39	2A/089B	(23)	2094	PUBLISHED
LARC	26-IN TRANSONIC BLOWDOWN	547	IS4	10*18*73 -	10*24*73	120/ 58	94	2A/089B	(30)	2146	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
LARC HIGH RE'S NUMBER HELIUM	100	FH1	11*15*72 -	01*01*73	160/ 80	200	TPS TILES	(15)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01, 1984

FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	REF.	MODEL	(ID)	DATAMAN NO.	DATAMAN STATUS
LARC 8-FT HIGH-TEMP STRUCTURE	655	SA2FA	7*24*73 -	08*07*73	60/176	176	SRB		(454)	2088	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
LARC	60-FT. VACUUM SPHERE	3289	OA99	3*26*74 -	4*12*74	50/ 52	14	3/139B	(21)	2172	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LERC	10X10-FT SUPERSONIC	035	SA6F	12* 3*73 - 01*16*74	120/208	0	SRB	(454)	2161	PUBLISHED
LERC	10X10-FT SUPERSONIC	038	IH34	5* 5*75 - 09*03*75	240/264	57	5/140C	(19)	2282	PUBLISHED
LERC	10X10-FT SUPERSONIC	041	IH39	9*22*76 - 04*14*77	240/226	163	5/140C	(19)	2435	PUBLISHED
LERC	10X10-FT SUPERSONIC	042	OA234	6* 7*77 - 08*11*77	80/ 80	63	ADS PROBES	(99)	2400	PUBLISHED
LERC	10X10-FT SUPERSONIC	044	IH83	1*25*78 - 03*10*78	200/102	41	5/140C	(19)	2440	PUBLISHED
LERC	10X10-FT SUPERSONIC	045	IH11	4* 1*78 - 04*18*78	80/ 64	0	5/140C	(84)	2428	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
LERC	SPACE POWER FACILITY	OH64	OH64	4*14*75 - 06*20*75	200/450	200	2A/089B	(25)	2288	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
MSFC 14-IN TRANSONIC	555	OA1	9*27*72 - 10*07*72	60/ 96	206	ATP	(1)	2005	PUBLISHED
MSFC 14-IN TRANSONIC	556	IA1A	10*10*72 - 10*19*72	56/ 84	179	ATP	(1)	2006	PUBLISHED
MSFC 14-IN TRANSONIC	545	IA1B	10*19*72 - 11*28*72	150/257	361	ATP	(1)	2010	PUBLISHED
MSFC 14-IN TRANSONIC	558	MA9F	11*29*72 - 12*07*72	74/ 75	132	ATP	(1)	2011	PUBLISHED
MSFC 14-IN TRANSONIC	554	SA1F	12* 9*72 - 12*23*72	160/144	200	PRR/SRB	(1)	2012	PUBLISHED
MSFC 14-IN TRANSONIC	565	SA3F	2*20*73 - 03*20*73	160/164	261	SRB	(449)	2025	PUBLISHED
MSFC 14-IN TRANSONIC	559	IS6B	3*20*73 - 05*27*73	50/ 50	70	2A/089B	(13)	2158	PUBLISHED
MSFC 14-IN TRANSONIC	568	OA47	3*28*73 - 04*05*73	116/116	245	2A/089B	(15)	2029	PUBLISHED
MSFC 14-IN TRANSONIC	566	IA31FA	4* 9*73 - 04*13*73	60/ 60	104	2A/089B	(13)	2026	PUBLISHED
MSFC 14-IN TRANSONIC	570	IA31FB	4*13*73 - 04*30*73	50/271	220	2A/089B	(13)	2028	PUBLISHED
MSFC 14-IN TRANSONIC	571	IA6	4*30*73 - 05*03*73	45/ 52	94	2A/089B	(13)	2039	PUBLISHED
MSFC 14-IN TRANSONIC	572	SA5F	5* 3*73 - 05*08*73	45/ 52	101	SRB	(449)	2051	PUBLISHED
MSFC 14-IN TRANSONIC	567	IA32F	5* 9*73 - 05*24*73	100/180	190	2A/089B	(13)	2027	PUBLISHED
MSFC 14-IN TRANSONIC	574	OA48	5*25*73 - 6*11*73	100/166	364	3/139B,W/CANS	(34)	2055	PUBLISHED
MSFC 14-IN TRANSONIC	573	IA31FC	6*21*73 - 07*09*73	32/ 51	145	2A/089B	(13)	2072	PUBLISHED
MSFC 14-IN TRANSONIC	579	IA37A	7*10*73 - 07*13*73	60/ 36	64	3A/139B	(34)	2063	PUBLISHED
MSFC 14-IN TRANSONIC	580	IA48	7*18*73 - 07*21*73	20/ 24	40	3A/139B	(34)	2063	PUBLISHED
MSFC 14-IN TRANSONIC	575	MA10F	7*23*73 - 09*12*73	160/305	0	OGIVE CYL	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	578	SA10F	9*13*73 - 10*01*73	112/128	200	SRB	(449)	2087	PUBLISHED
MSFC 14-IN TRANSONIC	582	IS6A	10* 2*73 - 10*11*73	80/ 74	126	2A/089B	(13)	2158	PUBLISHED
MSFC 14-IN TRANSONIC	584	IA52	10*11*73 - 10*17*73	16/ 28	27	3A/139B	(34)	2042	PUBLISHED
MSFC 14-IN TRANSONIC	585	IA37B	10*15*73 - 10*16*73	16/ 22	42	3A/139B	(34)	2093	PUBLISHED
MSFC 14-IN TRANSONIC	581	OA49	10*18*73 - 11*09*73	198/170	415	4/140A,B	(34)	2095	PUBLISHED
MSFC 14-IN TRANSONIC	589	IA62F	11*15*73 - 11*19*73	16/ 19	33	4/140A,B	(34)	2103	PUBLISHED
MSFC 14-IN TRANSONIC	590	SA26FA	11*19*73 - 12*11*73	63/ 63	100	SRB	(449)	2111	PUBLISHED
MSFC 14-IN TRANSONIC	588	IA53	12*20*73 - 01*04*74	40/ 36	45	2A/089B	(13)	2123	PUBLISHED

MSFC 14-IN TRANSONIC	586	MA11F	1* 3*74 - 01*18*74	80/102	0	OGIVE CYL	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	591	FA10	1* 7*74 - 01*14*74	40/ 56	0	JET PLUME SIM.	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	587	FA4	1*18*74 - 04*15*74	40/182	0	TITAN-3C	(459)	2142	PUBLISHED
MSFC 14-IN TRANSONIC	595	SA26FB	1*28*74 - 01*30*74	16/ 13	50	SRB	(449)	2111	PUBLISHED
MSFC 14-IN TRANSONIC	583	TA1F	2*19*74 - 03*05*74	56/ 64	0	ET	(459)	2145	PUBLISHED
MSFC 14-IN TRANSONIC	597	FA12	3* 6*74 - 03*10*74	24/ 24	0	CONE-CYLINDER	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	593	FA11	3*11*74 - 4*08*74	160/176	0	CONE-OGIVE-CYL	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	596	TA2F	4*29*74 - 09*23*74	104/104	0	ET	(460)	2165	PUBLISHED
MSFC 14-IN TRANSONIC	594	IA33	5* 9*74 - 07*21*74	256/264	270	5/140C	(74)	2174	PUBLISHED
MSFC 14-IN TRANSONIC	599	OA108	6*24*74 - 07*09*74	80/ 80	186	5/140C	(74)	2190	PUBLISHED
MSFC 14-IN TRANSONIC	607	OA131	9*11*74 - 09*26*74	80/ 96	109	5/140C	(74)	2232	PUBLISHED
MSFC 14-IN TRANSONIC	609	TA3F	9*27*74 - 10*11*74	64/ 80	0	ET	(470)	2208	PUBLISHED
MSFC 14-IN TRANSONIC	604	SA8F	10*18*74 - 12*10*74	160/250	0	SRB	(471)	2223	PUBLISHED
MSFC 14-IN TRANSONIC	610	IA71A-2	12*11*74 - 12*17*74	20/ 17	29	5/140C	(74)	2227	PUBLISHED
MSFC 14-IN TRANSONIC	610	IA71A-1	12*11*74 - 12*17*74	20/ 40	40	5/140C	(77)	2227	PUBLISHED
MSFC 14-IN TRANSONIC	610	IA71B-2	12*19*74 - 01*09*75	16/ 16	41	5/140C	(74)	2227	PUBLISHED
MSFC 14-IN TRANSONIC	610	IA71B-1	12*19*74 - 01*09*75	40/ 64	90	5 140C	(77)	2227	PUBLISHED
MSFC 14-IN TRANSONIC	600	FA14	1* 9*75 - 07*06*75	60/142	0	5/140C	(74)	2274	PUBLISHED
MSFC 14-IN TRANSONIC	616	SS14F	1*27*75 - 02*13*75	104/104	0	SRB	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	614	SS20F	2*15*75 - 02*21*75	40/ 44	0	SRB	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	611	SA30F	3* 3*75 - 03*13*75	80/ 72	185	SRB	(473)	2235	PUBLISHED
MSFC 14-IN TRANSONIC	603	SA28F-2	3*17*75 - 04*11*75	40/ 50	160	SRB	(469)	2244	PUBLISHED
MSFC 14-IN TRANSONIC	603	SA28F-1	3*17*75 - 04*11*75	80/102	200	SRB	(468)	2244	PUBLISHED
MSFC 14-IN TRANSONIC	623	SS15F	4*12*75 - 04*12*75	40/ 90	0	SRB	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	622	IA125-2	4*25*75 - 05*22*75	40/ 30	50	5/140C(74TS)	(77)	2253	PUBLISHED
MSFC 14-IN TRANSONIC	622	IA125-1	4*25*75 - 05*22*75	60/ 93	137	5/140C	(74)	2253	PUBLISHED
MSFC 14-IN TRANSONIC	626	SS19F	5*27*75 - 06*27*75	180/192	0	SRB	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	628	FA22A	7* 9*75 - 07*17*75	12/ 56	0	NOZZLE CALIB.	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	631	FA20A	7*18*75 - 07*22*75	24/ 24	0	NOZZLE CALIB.	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	612	FA13	8* 7*75 - 09*11*75	160/200	0	CONE CYLINDER	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	636	SS16F	9*22*75 - 10*20*75	120/158	0	SRB	(0)		UNASSIGNED

MSFC 14-IN TRANSONIC	633	FA21A	10*23*75 - 10*31*75	24/ 44	0	NOZZLE CALIB.	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	627	FA23A	11* 3*75 - 11*06*75	60/ 28	0	NOZZLE CALIB.	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	638	SS18F	11*14*75 - 12*08*75	80/131	0	SRB	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	620	SA14FA	12*23*75 - 03*19*76	140/144	200	SRB	(449)	2325	PUBLISHED
MSFC 14-IN TRANSONIC	640	SA14FB	1* 6*76 - 03*11*76	48/549	100	SRB	(486)	2310	PUBLISHED
MSFC 14-IN TRANSONIC	641	IA140A	6* 1*76 - 08*03*76	64/222	230	5/140C	(74)	2335	PUBLISHED
MSFC 14-IN TRANSONIC	643	TA6F	8* 4*76 - 08*20*76	40/ 80	0	ET INST.	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	645	SA21F	9*16*76 - 10*06*76	56/120	200	SRB	(486)	2345	PUBLISHED
MSFC 14-IN TRANSONIC	646	IA140B	10* 1*76 - 01*28*77	80/279	44	5/140C	(74)	2335	PUBLISHED
MSFC 14-IN TRANSONIC	632	FA15	1*31*77 - 05*01*77	104/320	0	OGIVE CYLINDER	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	630	FA19	5* 2*77 - 07*05*77	104/278	0	ACOUSTICS	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	648	SS30F	7*20*77 - 09*03*77	160/240	0	SRB	(0)		UNASSIGNED
MSFC 14-IN TRANSONIC	649	IA181	12*15*77 - 02*03*78	120/120	111	5/140C	(74)	2406	PUBLISHED
MSFC 14-IN TRANSONIC	652	FA25	4*15*78 - 08*01*78	200/294	0	5/140C	(74)	2437	PUBLISHED
MSFC 14-IN TRANSONIC	653	FA26	5* 1*78 - 06*01*78	80/ 80	0	5/140C	(74)		UNASSIGNED
MSFC 14-IN TRANSONIC	655	FA27	3*14*79 - 05*16*79	150/160	0	5/140C	(74)	2460	IN PROCESS
MSFC 14-IN TRANSONIC	657	FA29	4* 1*79 - 05*00*79	400/ 0	0	5/140C	(74)		UNASSIGNED
MSFC 14-IN TRANSONIC	656	FA28	8* 1*79 - 09*01*79	200/ 0	0	5/140C	(74)	2474	PUBLISHED

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
MSFC 32-IN LUDWIEG (HIGH RN)	031	MA12F	10* 1*73	- 02*04*74	80/328	0	OGIVE CYL	(0)		UNASSIGNED
MSFC 32-IN LUDWIEG (HIGH RN)	034	SA13F	9*30*74	- 06*17*75	100/100	0	SRB	(461)	2277	PUBLISHED
MSFC 32-IN LUDWIEG (HIGH RN)	038	FA21B	2*17*76	- 04*21*76	0/368	0	ROCKET MOTOR	(0)		UNASSIGNED
MSFC 32-IN LUDWIEG (HIGH RN)	039	SA31F	4*27*76	- 02*01*77	80/ 80	0	SRB	(487)	2369	PUBLISHED
MSFC 32-IN LUDWIEG (HIGH RN)	041	LA97A	4*18*77	- 08*15*77	80/400	116	L.E.VORTEX	(0)		UNASSIGNED
MSFC 32-IN LUDWIEG (HIGH RN)	041	LA97B	10* 1*78	- 06*01*79	80/400	0	L.E.VORTEX	(0)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01,1984

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
JPL	20-IN SUPERSONIC		FS8A	11* 1*73 -	11*08*73	40/ 32	0	PRR ASCENT	(0)		UNASSIGNED
JPL	20-IN SUPERSONIC		FS8B	8* 1*74 -	08*07*74	32/ 32	0	PRR ASCENT	(0)		UNASSIGNED
JPL	20-IN SUPERSONIC	702	MA21	8*15*75 -	09*04*75	80/ 92	50	5/140C	(34)		UNASSIGNED

APPENDIX A

TABLE A5 - WIND TUNNEL TESTING BY FACILITY - OTHER GOVERNMENT COMPLEXES

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01, 1984

FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	REF.	MODEL (ID)	DATAMAN NO.	DATAMAN STATUS
AEDC 1-FT TRANSONIC		OS47	11* 1*79	12*20*79	160/160	0	TPS	(110)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

AUG 01,1984

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
AEDC 16-FT TRANSONIC	470	IA105A	9* 2*77 -	11*20*77	290/281	885	5/140C	(47)	2398	PUBLISHED
AEDC 16-FT TRANSONIC	470	IA156A	10*28*77 -	11*10*77	96/124	575	VEH 102	(89)	2403	PUBLISHED
AEDC 16-FT TRANSONIC	431	OA232	2*17*78 -	03*01*78	80/ 80	281	ADS PROBES	(99)	2414	PUBLISHED
AEDC 16-FT TRANSONIC	507	OA129	7* 7*78 -	07*15*78	40/ 64	477	VEH102	(47)	2434	PUBLISHED
AEDC 16-FT TRANSONIC	517	IA182	9*19*78 -	09*20*78	12/ 24	87	5/140C	(47)	2439	PUBLISHED
AEDC 16-FT TRANSONIC	519	IA183	11*15*78 -	11*16*78	12/ 12	40	VEH 102	(89)	2444	PUBLISHED
AEDC 16-FT TRANSONIC	505	IA132	11*27*78 -	12*14*78	96/ 96	0	ET FORETANK	(68)	2449	PUBLISHED
AEDC 16-FT TRANSONIC	551	OS46A	3* 4*80 -	03*05*80	8/ 8	0	TPS	(109)	2505	PUBLISHED
AEDC 16-FT TRANSONIC	551	OS46B	3* 6*80 -	03*08*80	24/ 24	0	TPS	(108)	2505	PUBLISHED
AEDC 16-FT TRANSONIC	551	OS46C	4*17*80 -	04*18*80	8/ 8	0	TPS	(109)	2505	PUBLISHED
AEDC 16-FT TRANSONIC	574	OA253	7* 1*80 -	07*08*80	80/ 80	139	5/140C	(84)	2486	PUBLISHED
AEDC 16-FT TRANSONIC	551	OS46D	8* 0*80 -	09*00*80	8/ 8	0	TPS	(108)	2505	PUBLISHED
AEDC 16-FT TRANSONIC	551	OS46E	9*15*80 -	10*20*80	8/ 8	0	TPS	(108)	2505	PUBLISHED
AEDC 16-FT TRANSONIC	551	OS46F	10* 1*80 -	10*02*80	8/ 8	0	TPS	(108)	2505	PUBLISHED
AEDC 16-FT TRANSONIC	556	OS49	1*28*81 -	02*04*81	40/ 44	0	TPS	(111)	2483	PUBLISHED
AEDC 16-FT TRANSONIC	594	MA34	3*12*81 -	03*20*81	40/ 60	0	ADS PROBES	(99)	2497	IN PROCESS
AEDC 16-FT TRANSONIC	608	OS56	8*26*81 -	08*27*81	8/ 8	0	TPS TILE	(108)	2489	PUBLISHED
AEDC 16-FT TRANSONIC	551	OS46G	12*10*81 -	12*11*81	8/ 17	0	TPS	(108)	2505	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY   AUG 01,1984

FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
AEDC 4-FT TRANSONIC	390	FA22B	7*21*75 - 07*25*75	30/ 35	0	OGIVE CYLINDER	(0)		UNASSIGNED
AEDC 4-FT TRANSONIC	409	FA20B	10* 3*75 - 10*13*75	60/ 52	0	TRIPLE BODY	(0)		UNASSIGNED
AEDC 4-FT TRANSONIC	445	SA16F	5* 5*76 - 05*06*76	8/ 8	9	SRB	(486)	2334	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
AEDC A / SUPERSONIC	323	IA13	7* 5*73 - 07*17*73	40/ 39	762	3/139B	(32)	2062	PUBLISHED
AEDC A / SUPERSONIC	422	IA57	11*20*73 - 11*20*73	10/ 9	10	3/139,089B	(32)	2112	PUBLISHED
AEDC A / SUPERSONIC	422	IA61A	1*30*74 - 01*31*74	10/ 10	88	3/139,089B	(32)	2143	PUBLISHED
AEDC A / SUPERSONIC	21AA	IA61B	2*26*74 - 02*26*74	8/ 8	9	3/139,089B	(52)	2226	PUBLISHED
AEDC A / SUPERSONIC	60A	IA87	7*18*74 - 07*20*74	24/ 23	90	3/139B	(52)	2192	PUBLISHED
AEDC A / SUPERSONIC	71A	OA115A	7*29*74 - 07*31*74	24/ 28	82	4/140A,B(MOD)	(49)	2198	PUBLISHED
AEDC A / SUPERSONIC	A3A	IA111	3*21*75 - 03*28*75	36/ 33	***	3/139B	(52)	2242	PUBLISHED
AEDC A / SUPERSONIC	4A	IH41A	3*31*75 - 5*21*75	48/ 57	318	5/140C	(60)	2240	PUBLISHED
AEDC A / SUPERSONIC	F5A	SS17F	4* 4*75 - 04*05*75	12/ 12	0	SRB	(0)		UNASSIGNED
AEDC A / SUPERSONIC	E1A	FH13	9*22*75 - 09*25*75	24/ 40	0	ET/SPIKE	(0)	2276	PUBLISHED
AEDC A / SUPERSONIC	4A	IH41B	12*11*75 - 01*09*76	78/ 80	300	5/140C	(60)	2295	PUBLISHED
AEDC A / SUPERSONIC	EOA	SH15F	12*29*75 - 02*20*76	12/100	0	SRB	(0)		UNASSIGNED
AEDC A / SUPERSONIC	F9A	SS22F	1* 1*76 - 01*15*76	64/ 45	0	SRB	(0)		UNASSIGNED
AEDC A / SUPERSONIC	J3A	IH47	3* 8*76 - 03*19*76	40/ 31	178	5/140C	(60)	2312	PUBLISHED
AEDC A / SUPERSONIC	E6A	SH16F	3*10*76 - 04*19*76	12/ 8	0	SRB	(0)		UNASSIGNED
AEDC A / SUPERSONIC	425	IA40	6*23*76 - 06*29*76	26/ 41	346	5/140C	(75)	2293	PUBLISHED
AEDC A / SUPERSONIC	K1A	IA142	8*11*76 - 08*18*76	78/ 64	***	5/140C	(75)	2346	PUBLISHED
AEDC A / SUPERSONIC	K8A	MA28	9*29*76 - 09*29*76	7/ 1	0	2A/089B	(6)		UNASSIGNED
AEDC A / SUPERSONIC	P8A	IA143	11* 8*76 - 11*13*76	65/ 58	***	5/140C	(75)	2354	PUBLISHED
AEDC A / SUPERSONIC	K2A	IH72	1* 3*77 - 01*10*77	60/ 56	0	5/140C	(60)	2372	PUBLISHED
AEDC A / SUPERSONIC	P5A	OA209	3*21*78 - 03*30*78	65/ 69	324	VEH. 102	(105)	2415	PUBLISHED
AEDC A / SUPERSONIC	W5	IH85	4*19*78 - 04*26*78	60/ 65	337	5/140C	(60)	2431	PUBLISHED
AEDC A / SUPERSONIC	420	FH15	5* 1*78 - 05*05*78	52/ 52	0	ET/SPIKE	(0)	2422	PUBLISHED
AEDC A / SUPERSONIC	B67	IH102-3	5* 1*79 - 06*01*79	10/ 10	0	5/140C	(83)	2464	PUBLISHED
AEDC A / SUPERSONIC	B67	IH102-2	5* 1*79 - 06*01*79	12/ 12	0	5/140C	(56)	2464	PUBLISHED
AEDC A / SUPERSONIC	B67	IH102-1	5* 1*79 - 06*01*79	26/ 26	0	5/140C	(60)	2464	PUBLISHED

AEDC A / SUPERSONIC A1G IA193 2*26*82 - 04*31*82 72/720 O VEH 102 (72) UNASSIGNED

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PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
AEDC B	HYPERSONIC	288	OH3A	6*28*73 - 06*30*73	40/ 16	36	3/139B	(21)	2100	PUBLISHED
AEDC B	HYPERSONIC	289	OH3B	7* 9*73 - 07*11*73	40/ 23	147	3/139B	(21)	2100	PUBLISHED
AEDC B	HYPERSONIC	353	OH9	9*13*73 - 09*21*73	16/ 16	61	3/139	(29)	2251	PUBLISHED
AEDC B	HYPERSONIC	352	OH4C	9*26*73 - 09*26*73	8/ 8	60	3/139B	(21)	2225	PUBLISHED
AEDC B	HYPERSONIC	352	OH4B	9*29*73 - 10*04*73	48/ 38	224	3/139	(22)	2099	PUBLISHED
AEDC B	HYPERSONIC	352	OH4A	11*12*73 - 12*05*73	20/ 20	57	3/139	(29)	2154	PUBLISHED
AEDC B	HYPERSONIC	474	OA77	11*27*73 - 12*01*73	40/ 32	124	4/140A,B	(49)	2134	PUBLISHED
AEDC B	HYPERSONIC	422	IA17A	3* 6*74 - 03*15*74	40/ 45	997	3/139B	(52)	2156	PUBLISHED
AEDC B	HYPERSONIC	422	IA17B	3*18*74 - 03*19*74	8/ 8	13	3/139B	(52)	2230	PUBLISHED
AEDC B	HYPERSONIC	526	OH50A	3*29*74 - 04*11*74	8/ 16	66	5/140C	(82)	2285	PUBLISHED
AEDC B	HYPERSONIC	525	OH49A	4* 3*74 - 04*06*74	216/ 17	87	3/139B	(22)	2355	PUBLISHED
AEDC B	HYPERSONIC	524	OH52	5* 6*74 - 05*15*74	16/ 16	32	3/139B	(29)	2330	PUBLISHED
AEDC B	HYPERSONIC	550	LA42A	6*25*74 - 06*25*74	16/ 8	3	089B	(0)	2132	PUBLISHED
AEDC B	HYPERSONIC	57A	OH49B	7* 2*74 - 07*12*74	72/ 67	454	4/140B	([illegible])	2222	PUBLISHED
AEDC B	HYPERSONIC	58A	OH50B	7*12*74 - 07*17*74	36/ 27	220	5/140C	(83)	2358	PUBLISHED
AEDC B	HYPERSONIC	48A	LA42B	7*27*74 - 07*27*74	16/ 12	7	089B	(0)	2132	PUBLISHED
AEDC B	HYPERSONIC	71A	OA79	8* 1*74 - 08*03*74	24/ 23	79	4/140A,B(MOD)	(49)	2196	PUBLISHED
AEDC B	HYPERSONIC	83A	OH25A	8*21*74 - 08*22*74	12/ 12	82	3/139B	(21)	2252	PUBLISHED
AEDC B	HYPERSONIC	82A	OH54A	10* 4*74 - 10* 8*74	36/ 32	117	5/140C	(82)	2301	PUBLISHED
AEDC B	HYPERSONIC	74A	OH39A	11*21*74 - 11*28*74	84/ 59	622	5/140C	(60)	2241	PUBLISHED
AEDC B	HYPERSONIC	74A	OH39B	1* 8*75 - 01*09*75	12/ 13	80	5/140C	(60)	2241	PUBLISHED
AEDC B	HYPERSONIC	83A	OH25B	1*30*75 - 02*03*75	24/ 23	153	5/140C	(56)	2366	PUBLISHED
AEDC B	HYPERSONIC	87A	OH60	5*12*75 - 05*12*75	12/ 12	139	5/140C	(83)	2356	PUBLISHED
AEDC B	HYPERSONIC	87A	OH74	6* 3*75 - 06*12*75	12/ 12	0	5/140C	(56)	2263	PUBLISHED
AEDC B	HYPERSONIC	82A	OH54B	7*21*75 - 07*25*75	48/ 52	124	5/140C	(82)	2342	PUBLISHED
AEDC B	HYPERSONIC	C4A	IA114	8*18*75 - 08*22*75	42/ 56	100	5/140C	(52)	2272	PUBLISHED

AEDC B / HYPERSONIC	82A	OH54C	8*26*75 - 09*02*75	48/ 48	120	5/140C	(82)	2342	PUBLISHED
AEDC B / HYPERSONIC	E3A	OH75	9* 2*75 - 09*03*75	14/ 13	44	5/140C	(82)	2303	PUBLISHED
AEDC B / HYPERSONIC	D5A	MH2	9* 3*75 - 01*23*76	16/ 11	22	4/140B	(29)		UNASSIGNED
AEDC B / HYPERSONIC	E9A	OH69	11*14*75 - 12*11*75	84/ 87	246	5/140C	(82)	2321	PUBLISHED
AEDC B / HYPERSONIC	D8A	OA169	3*26*76 - 04*09*76	12/ 43	200	5/140C	(70)	2320	PUBLISHED
AEDC B / HYPERSONIC	59A	IA22	5* 3*76 - 05*08*76	52/ 49	750	5/140C	(70)	2327	PUBLISHED
AEDC B / HYPERSONIC	J7A	OH98A	6*17*76 - 06*23*76	43/ 44	284	5/140C	(60)	2340	PUBLISHED
AEDC B / HYPERSONIC	J74	OH98B	7*26*76 - 07*26*76	20/ 13	98	5/140C	(60)	2340	PUBLISHED
AEDC B / HYPERSONIC	K3A	OH57A	10* 6*76 - 20*06*76	13/ 11	40	140C	(92)	2367	PUBLISHED
AEDC B / HYPERSONIC	K7A	MA29	10*14*76 - 10*14*46	7/ 7	0	SEMISPAN	(0)	2451	PUBLISHED
AEDC B / HYPERSONIC	K3A	OH57B	12* 4*76 - 12*05*76	26/ 34	14	140C	(92)	2367	PUBLISHED
AEDC B / HYPERSONIC	R4A	OH84A-2	4*20*77 - 04*21*77	5/ 9	16	5/140C	(83)	2388	PUBLISHED
AEDC B / HYPERSONIC	R4A	OH84A-1	4*20*77 - 04*21*77	20/ 16	81	5/140C	(60)	2388	PUBLISHED
AEDC B / HYPERSONIC	TOA	IA148	4*27*77 - 05*03*77	52/ 52	272	5/140C	(70)	2384	PUBLISHED
AEDC B / HYPERSONIC	R3A	OH56	12* 6*77 - 12*10*77	48/ 36	255	WING TIP SEAL	(91)	2410	PUBLISHED
AEDC B / HYPERSONIC	V2C	OH103A	2*20*78 - 02*21*78	12/ 8	72	VEH. 5 F'BODY	(83)	2420	PUBLISHED
AEDC B / HYPERSONIC	P4A	OH90	3* 2*78 - 03*11*78	48/ 64	162	ELEV/ELEV	(94)	2451	PUBLISHED
AEDC B / HYPERSONIC	P5A	OA208	3*30*78 - 04*06*78	52/ 47	183	VEH. 102	(105)	2416	PUBLISHED
AEDC B / HYPERSONIC	B65	OH102A	10*25*78 - 11*29*78	8/ 13	0	5/140C	(56)	2455	PUBLISHED
AEDC B / HYPERSONIC	B67	OH84B	5* 0*79 - 06*00*79	72/ 72	0	5/140C	(60)	2464	PUBLISHED
AEDC B / HYPERSONIC	B67	OH105A	5*15*79 - 06*20*79	24/ 24	0	5/140C	(60)	2464	PUBLISHED
AEDC B / HYPERSONIC	G9	OH109	10*27*80 - 11*24*80	48/ 40	0	5/140C	(56)	2490	PUBLISHED
AEDC B / HYPERSONIC	BHQ	OA258	11*25*80 - 01*06*81	48/128	541	VEH 102	(106)	2491	PUBLISHED
AEDC B / HYPERSONIC	B17	OH107	1* 7*81 - 01*08*81	12/ 12	0	ELEV/ELEV	(94)	2492	PUBLISHED
AEDC B / HYPERSONIC	14	OA259	2*16*81 - 02*20*81	16/ 40	137	VEH 102	(72)	2493	PUBLISHED
AEDC B / HYPERSONIC	1C	OH111	9*24*81 - 09*30*81	32/ 32	0	5/140C	(60)	2496	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
AEDC C / HYPERSONIC	474	OA78	12* 3*73 - 12*04*73	20/ 16	56	4/140A,B	(49)	2134	PUBLISHED

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PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	REF.	MODEL	(ID)	DATAMAN NO.	DATAMAN STATUS
AEDC D / HYPERSONIC		V2C	OH103B	4*27*78 -	04*28*78	24/ 12	53	5/140C		(60)	2427	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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WIND TUNNEL FACILITY	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
NSWC HYPERSONIC LAB (#9)	1310	OA171	6* 5*78 - 06*22*78	180/180	35	VEH. 102	(105)	2433	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	REF.	MODEL	(ID)	DATAMAN NO.	STATUS
NSWC HYPERVELOCITY LAB (#8A)	1275	LA79	11*28*75 - 12*11*75	64/ 64	8	140C		(0)	2291	IN PROCESS

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TABLE A6 - WIND TUNNEL TESTING BY FACILITY - PRIVATE FACILITY COMPLEXES

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
CALSPAN 32-IN LUDWIEG	181	IH5	1*21*74 - 07*22*74	120/105	106	2A/089B	(19)	2308	PUBLISHED
CALSPAN 32-IN LUDWIEG	033	SA29F	8* 8*74 - 09*18*74	120/120	0	SRB FORE BODY	(467)	2207	PUBLISHED
CALSPAN 32-IN LUDWIEG	100	IH75	10* 3*77 - 12*12*77	200/320	41	5/140C	(19)	2453	PUBLISHED

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
CALSPAN	8-FT TRANSONIC	053	IA36	6*15*73 - 06*22*73	60/ 80	120	2A/089(MOD)	(14)	2064	PUBLISHED
CALSPAN	8-FT TRANSONIC	103	LA70	7*28*75 - 08*06*75	38/ 60	299	140C/REMOTE ELE	(44)	2269	PUBLISHED
CALSPAN	8-FT TRANSONIC	111	LA82	8* 8*76 - 08*19*76	30/ 32	66	SUPPORT TARES	(20?)	2374	PUBLISHED
CALSPAN	8-FT TRANSONIC	113	LA103	3*25*77 - 04*04*77	80/ 76	88	SUPPORT TARES	(0)	2374	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED.		TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
CALSPAN	HYPERSONIC SHOCK	100	IH21	10*29*73	-	12*13*73	80/14[illegible]	31	3/139	(37)	2164	PUBLISHED
CALSPAN	HYPERSONIC SHOCK	100	OH12	10*29*73	-	12*13*73	80/145	32	3/139	(37)	2164	PUBLISHED
CALSPAN	HYPERSONIC SHOCK	184-	OA113	8*10*74	-	10*04*74	24/336	108	4/140A,B	(51)	2234	PUBLISHED
CALSPAN	HYPERSONIC SHOCK	120	IH33A	10*14*74	-	10*18*74	32/ 32	1[illegible]	5/140C	(37)	2249	PUBLISHED
CALSPAN	HYPERSONIC SHOCK	737	OA93	11*18*74	-	11*23*74	80/152	[illegible]	4/140A,B	(51)	2238	PUBLISHED
CALSPAN	HYPERSONIC SHOCK	131	IH33B	12* 5*74	-	12*19*74	48/ 80	24	5/140C	(37)	2249	PUBLISHED
CALSPAN	HYPERSONIC SHOCK	000	TH2F	6* 1*75	-	06*05*75	80/ 32	0	ET	(0)		UNASSIGNED
CALSPAN	HYPERSONIC SHOCK	189	IH43	12*17*75	-	02*23*76	120/250	60	5/140C	(59)	2319	PUBLISHED
CALSPAN	HYPERSONIC SHOCK	131	OH66	8*30*76	-	10*17*76	120/120	30	5/140C	(66)	2359	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	REF. MODEL	(ID)	DATAMAN NO.	STATUS
GRUMMAN - LOW SPEED	324	MA18	6* 5*74 - 06*22*74	200/200	254	GULFSTREAM 2	(0)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LOCKHEED (CA) - LOW SPEED	363	MA16	10* 3*73 - 10*12*73	40/ 56	106	089B/C-5A	(2)		UNASSIGNED
LOCKHEED (CA) - LOW SPEED	365	CA103	11*26*73 - 11*28*73	24/ 24	45	089B/C-5A	(2)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LOCKHEED (GA) - V/STOL	120	CA104	12*13*73 - 01*21*74	160/165	208	089B/C-5A	(2)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LOCKHEED (GA) - LOW SPEED		CA1	5*30*74 - 06*04*74	40/ 56	50	ET/C-5A	(399)		UNASSIGNED
LOCKHEED (GA) - LOW SPEED		CA2-1	6* 4*74 - 06*10*74	120/ 80	100	4/140A,B/C-5A	(43)		UNASSIGNED
LOCKHEED (GA) - LOW SPEED		CA2-2	6*11*74 - 06*23*74	40/ 40	64	ET/C-5A	(399)		UNASSIGNED
LOCKHEED (GA) - LOW SPEED	190	SA38F	9*23*76 - 09*27*76	32/ 30	0	SRB	(0)		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
LTV 15X20-FT LOW SPEED	407	MA1	8*25*72 - 09*06*72	80/ 80	120	JSC 040A ORB.	(95)	2004	PUBLISHED
LTV 15X20-FT LOW SPEED	422	MA14	4*23*73 - 05*02*73	80/ 62	103	2A/089B(CAN)	(95)	2283	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
LTV 4X4-FT	SUPERSONIC	458	IA4	11* 2*72 - 11*17*72	80/ 75	62	PRE-ATP/001	(9)	2015	PUBLISHED
LTV 4X4-FT	SUPERSONIC	488	OA84	12*10*73 - 12*14*73	80/115	207	4/140A,B	(49)	2037	PUBLISHED
LTV 4X4-FT	SUPERSONIC	498	LA28	6*17*74 - 06*20*74	40/ 40	31	140A,B ORB	(0)	2280	PUBLISHED
LTV 4X4-FT	SUPERSONIC	512	LA58	9*30*74 - 10*04*74	49/ 80	72	140A,B	(42)	2215	PUBLISHED
LTV 4X4-FT	SUPERSONIC	552	LA67	6*20*75 - 07*02*75	40/120	131	140C/REMOTE ELE	(44)	2266	PUBLISHED
LTV 4X4-FT	SUPERSONIC	559	CA26	8* 4*75 - 08*15*75	94/ 95	131	140C(MOD)/747	(48)	2273	PUBLISHED
LTV 4X4-FT	SUPERSONIC	573	LA76	2*25*76 - 03*06*76	48/128	141	140C/REMOTE ELE	(44)	2305	PUBLISHED
LTV 4X4-FT	SUPERSONIC	742	LA144	7*28*80 - C8*01*80	80/138	198	VEH 102	(106)	2484	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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WIND TUNNEL FACILITY	NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
THE BOEING CO. - V/STOL	132	CA92	11*27*73 -	12*04*73	80/ 97	114	089B/747	(2)		UNASSIGNED

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
THE BOEING CO.	- TRANSONIC	1431	CA5	9*20*74 - 09*30*74	144/181	520	140A,B/747	(45)	2211	PUBLISHED
THE BOEING CO.	- TRANSONIC	1431	CA20	10* 9*74 - 10*15*74	115/115	288	140A,B/747	(45)	2217	PUBLISHED
THE BOEING CO.	- TRANSONIC	1472	CA6	5*20*75 - 06*06*75	200/265	5C9	140A,B/747	(45)	2262	PUBLISHED
THE BOEING CO.	- TRANSONIC	1474	CS2	6* 9*75 - 06*16*75	95/ 95	165	140A,B/747	(45)		UNASSIGNED
THE BOEING CO.	- TRANSONIC	1477	CA9	6*25*75 - 07*14*75	320/302	85	4/140A,B/747	(47)	2268	PUBLISHED
THE BOEING CO.	- TRANSONIC	1490	CS4	9*29*75 - 10*02*75	40/ 64	95	140A,B/747	(45)	2341	PUBLISHED
THE BOEING CO.	- TRANSONIC	1493	CS5	11* 3*75 - 11*05*75	24/ 33	192	140A,B/747	(45)	2341	PUBLISHED
THE BOEING CO.	- TRANSONIC	1496	CA14	11*13*75 - 12*02*75	160/236	850	140A,B/747	(45)	2307	PUBLISHED

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TABLE A7 - WIND TUNNEL TESTING BY FACILITY - SPACE SHUTTLE PRIME CONTRACTOR COMPLEX

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
RI 7X11-FT LOW SPEED	689	OA2	9*25*72 - 10*03*72	80/ 69	183	ATP	(2)	2016	PUBLISHED
RI 7X11-FT LOW SPEED	690	OA5	10*11*72 - 10*19*72	60/ 65	88	ATP-MODIF	(2)	2017	PUBLISHED
RI 7X11-FT LOW SPEED	693	IA?	11* 3*72 - 11*16*72	24/ 41	53	PRE-ATP/001	(10)	2018	PUBLISHED
RI 7X11-FT LOW SPEED	694	OA6	11*16*72 - 12*06*72	60/177	218	PRR	(2)	2019	PUBLISHED
RI 7X11-FT LOW SPEED	696	OA9	12*18*72 - 01*09*73	60/158	192	2A/089B	(2)	2020	PUBLISHED
RI 7X11-FT LOW SPEED	698	OA10	1*30*73 - 02*16*73	120/109	300	2A/089B	(2)	2022	PUBLISHED
RI 7X11-FT LOW SPEED	699	OA45	2*21*73 - 02*28*73	80/ 86	171	2A/089B	(2)	2021	PUBLISHED
RI 7X11-FT LOW SPEED	700	OA14	2*28*73 - 03*15*73	100/151	196	2A/089B	(2)	2030	PUBLISHED
RI 7X11-FT LOW SPEED	701	OA16	3*19*73 - 04*17*73	130/320	475	2A/089B	(2)	2038	PUBLISHED
RI 7X11-FT LOW SPEED	704	OA18	5* 8*73 - 05*17*73	100/114	189	3/139B	(43)	2045	PUBLISHED
RI 7X11-FT LOW SPEED	705	OA21A	5*21*73 - 06*04*73	100/ 72	348	3/139B	(43)	2053	PUBLISHED
RI 7X11-FT LOW SPEED	705	OA21B	6*21*73 - 06*25*73	40/ 55	99	3/139B W/CANS	(43)	2053	PUBLISHED
RI 7X11-FT LOW SPEED	708	OA71A	7*27*73 - 08*03*73	50/ 62	52	2A/089B	(2)	2068	PUBLISHED
RI 7X11-FT LOW SPEED	709	OA57A	8* 6*73 - 8*17*73	100/103	61	2A/089B	(2)	2074	PUBLISHED
RI 7X11-FT LOW SPEED	711	OA69	8*28*73 - 09*01*73	80/ 71	205	3/139B	(43)	2081	PUBLISHED
RI 7X11-FT LOW SPEED	712	OA71C	9* 4*73 - 09*14*73	100/139	71	3/139B	(43)	2086	PUBLISHED
RI 7X11-FT LOW SPEED	713	OA57B	9*15*73 - 09*17*73	40/123	72	2A/089B	(2)	2080	PUBLISHED
RI 7X11-FT LOW SPEED	715	OA62A	10* 5*73 - 10*23*73	120/195	98	4/140A,B	(43)	2097	PUBLISHED
RI 7X11-FT LOW SPEED	716	OA86	10*26*73 - 11*09*73	80/174	331	4/140A,B	(43)	2114	PUBLISHED
RI 7X11-FT LOW SPEED	717	OA62B	11*13*73 - 12*06*73	100/240	448	4/140A,B	(43)	2104	PUBLISHED
RI 7X11-FT LOW SPEED	719	OA37	1* 7*74 - 01*25*74	80/103	112	4/140A,B	(47)	2140	PUBLISHED
RI 7X11-FT LOW SPEED	721	OA110	3*15*74 - 03*20*74	80/ 48	85	4/140A,B	(16)	2155	PUBLISHED
RI 7X11-FT LOW SPEED	724	OA118	4*24*74 - 04*25*74	48/ 40	54	4/140A,B	(43)	2139	PUBLISHED
RI 7X11-FT LOW SPEED	726	OA119A	6*17*74 - 06*25*74	20/ 45	45	4/140A,B	(16)	2187	PUBLISHED
RI 7X11-FT LOW SPEED	730	OA119B	8*22*74 - 09*06*74	60/100	213	4/140A,B	(16)	2203	PUBLISHED
RI 7X11-FT LOW SPEED	731	OA123	9* 6*74 - 09*10*74	40/ 47	41	4/140A,B (ALT)	(43)	2202	PUBLISHED

RI	7X11-FT LOW SPEED	736	OA124	10*14*74 - 10*23*74	60/ 60	127	4/140A,B	(43)	2209	PUBLISHED
RI	7X11-FT LOW SPEED	737	OA143	11* 6*74 - 11*11*74	40/ 55	60	4/140A,B	(16)	2221	PUBLISHED
RI	7X11-FT LOW SPEED	751	OA163A	11*24*75 - 12*09*75	160/144	215	4/140A,B	(16)	2289	PUBLISHED
RI	7X11-FT LOW SPEED	752	OA172	12*15*75 - 01*13*76	120/210	122	4/140A,B(ALT)	(43)	2294	PUBLISHED
RI	7X11-FT LOW SPEED	754	OA176	3*29*76 - 04*15*76	60/ 83	113	4/140A,B(ALT)	(43)	2314	PUBLISHED
RI	7X11-FT LOW SPEED	759	OA236	5*28*76 - 06*02*76	10/ 37	204	ADS PROBES	(99)	2337	PUBLISHED
RI	7X11-FT LOW SPEED	757	OA228	5*29*76 - 05*01*76	16/ 23	45	VEH 102 (ADS)	(57)	2322	PUBLISHED
RI	7X11-FT LOW SPEED	764	OA238	10*25*76 - 11*08*76	24/ 48	57	ADS PROBES	(99)	2351	PUBLISHED
RI	7X11-FT LOW SPEED	776	OA223	11*20*76 - 11*30*76	40/ 88	13	VEH 102	(39)	2402	PUBLISHED
RI	7X11-FT LOW SPEED	788	OA163B	12*21*76 - 12*23*76	35/ 35	99	4/140A,B	(16)	2361	PUBLISHED
RI	7X11-FT LOW SPEED	775	OA250	7* 1*77 - 07*07*77	32/ 34	23	140C(ALT)	(45)	2392	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER FACILITY

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FACILITY	WIND TUNNEL NO.	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	DATAMAN STATUS
RI 7-FT TRISONIC	276	OA68	6*20*73 -	6*29*73	60/ 60	44	3A/140A	(49)	2061	PUBLISHED
RI 7-FT TRISONIC	278	OA91	10*26*73 -	11*01*73	40/ 40	38	4/140A/B	(49)	2116	PUBLISHED
RI 7-FT TRISONIC	280	IA69	1*10*74 -	01*14*75	24/ 25	14	4/140A,B	(67)	2122	PUBLISHED
RI 7-FT TRISONIC	281	IA68	1*18*74 -	01*29*74	32/ 36	34	2A/089B	(13)	2144	PUBLISHED
RI 7-FT TRISONIC	282	IA70	5* 3*74 -	05*24*74	80/161	173	4/140A,B	(43)	2175	PUBLISHED
RI 7-FT TRISONIC	297	IA141	3*31*76 -	04*05*76	30/ 30	37	5/140C	(72)	2315	PUBLISHED

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TABLE A8 - WIND TUNNEL TESTING BY FACILITY - UNIVERSITY FACILITIES

PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
TEXAS A+M	7X10-FT LOW SPEED	MA8	MA8	12*15*72 - 01*27*73	40/ 40	40	JSC 040A ORB.	(95)		UNASSIGNED
TEXAS A+M	7X10-FT LOW SPEED	7513	MA24	7* 9*75 - 08*11*75	24/176	200	2A/089B(MOD)	(2)		UNASSIGNED
TEXAS A+M	7X10-FT LOW SPEED	7515	CA16	8*23*75 - 09*05*75	72/ 84	60	140A,B/747	(45)		UNASSIGNED

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PHASE C/D WIND TUNNEL TESTING PER FACILITY

FACILITY	WIND TUNNEL	NO.	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	MODEL REF.	(ID)	DATAMAN NO.	STATUS
UNIV. OF WASH.	LOW SPEED	1128	CA4	5*28*74 - 06*07*74	64/120	100	4/140A,B/747	(43)		UNASSIGNED
UNIV. OF WASH.	LOW SPEED	1136	CA3	8*15*74 - 08*30*74	120/131	194	4/140A,B/747	(43)	2201	PUBLISHED
UNIV. OF WASH.	LOW SPEED	1146	CA11	2*12*75 - 02*20*75	100/116	120	ET/747	(0)	2236	PUBLISHED
UNIV. OF WASH.	LOW SPEED	1160	CS1	6* 9*75 - 06*13*75	80/ 95	165	140A,B/747	(8)		UNASSIGNED
UNIV. OF WASH.	LOW SPEED	1170	CS3	9*12*75 - 09*15*75	40/ 80	129	140A,B/747	(8)	2338	PUBLISHED
UNIV. OF WASH.	LOW SPEED	1173	CA15A	10*16*75 - 11*01*75	240/239	379	4/140A,B/747	(43)	2347	PUBLISHED
UNIV. OF WASH.	LOW SPEED	1178	CA15B	11*19*75 - 11*26*75	75/110	93	4/140A,B/747	(43)	2348	PUBLISHED
UNIV. OF WASH.	LOW SPEED	1184	CA17	6*21*76 - 07*02*76	152/152	261	4/140A,B/747	(43)	2349	PUBLISHED

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TABLE A9 - WIND TUNNEL TESTING BY MODEL

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 1

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
ATP	.0040	MA2	9*18*72 - 11*06*72	40/ 80	31	LARC 22-IN HELIUM	409	2003	PUBLISHED
ATP	.0040	OA1	9*27*72 - 10*07*72	60/ 96	206	MSFC 14-IN TRANSONIC	555	2005	PUBLISHED
ATP	.0040	IA1A	10*10*72 - 10*19*72	5[illegible]/ 84	179	MSFC 14-IN TRANSONIC	556	2006	PUBLISHED
ATP	.0040	IA1B	10*19*72 - 11*28*72	150/257	361	MSFC 14-IN TRANSONIC	545	2010	PUBLISHED
ATP	.0040	MA9F	11*29*72 - 12*07*72	[illegible]4/ 75	132	MSFC 14-IN TRANSONIC	558	2011	PUBLISHED
PRR/SRB	.0040	SA1F	12* 9*72 - 12*23*72	160/144	200	MSFC 14-IN TRANSONIC	554	2012	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 2

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
ATP	.0405	OA2	9*25*72 - 10*03*72	80/ 69	183	RI 7X11-FT LOW SPEED	689	2016	PUBLISHED
ATP-MODIF	.0405	OA5	10*11*72 - 10*19*72	60/ 65	88	RI 7X11-FT LOW SPEED	690	2017	PUBLISHED
PRR	.0405	OA6	11*16*72 - 12*06*72	60/177	218	RI 7X11-FT LOW SPEED	694	2019	PUBLISHED
2A/089B	.0405	OA9	12*18*72 - 01*09*73	60/158	192	RI 7X11-FT LOW SPEED	696	2020	PUBLISHED
2A/089B	.0405	OA10	1*30*73 - 02*16*73	120/109	300	RI 7X11-FT LOW SPEED	698	2022	PUBLISHED
2A/089B	.0405	OA45	2*21*73 - 02*28*73	80/ 86	171	RI 7X11-FT LOW SPEED	699	2021	PUBLISHED
2A/089B	.0405	OA14	2*28*73 - 03*15*73	100/151	196	RI 7X11-FT LOW SPEED	700	2030	PUBLISHED
2A/089B	.0405	OA16	3*19*73 - 04*17*73	130/320	475	RI 7X11-FT LOW SPEED	701	2038	PUBLISHED
2A/089B	.0405	OA71A	7*27*73 - 08*03*73	50/ 62	52	RI 7X11-FT LOW SPEED	708	2068	PUBLISHED
2A/089B	.0405	OA57A	8* 6*73 - 8*17*73	100/103	61	RI 7X11-FT LOW SPEED	709	2074	PUBLISHED
2A/089B	.0405	OA57B	9*15*73 - 09*17*73	40/123	72	RI 7X11-FT LOW SPEED	713	2080	PUBLISHED
089B/C-5A	.0405	MA16	10* 3*73 - 10*12*73	40/ 56	106	LOCKHEED (CA) - LOW SPEED	363		UNASSIGNED
089B/C-5A	.0400	CA103	11*26*73 - 11*28*73	24/ 24	45	LOCKHEED (CA) - LOW SPEED	365		UNASSIGNED
089B/747	.0405	CA92	11*27*73 - 12*04*73	80/ 97	114	THE BOEING CO. - V/STOL	132		UNASSIGNED
089B/C-5A	.0405	CA104	12*13*73 - 01*21*74	160/165	208	LOCKHEED (GA) - V/STOL	120		UNASSIGNED
2A/089B(MOD)	.0405	MA24	7* 9*75 - 08*11*75	24/176	200	TEXAS A+M 7X10-FT LOW SPEED	7513		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 3

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
PRE-ATP/001	.0060	OH1A-1	9*19*72 - 09*26*72	10/ 10	130	LARC MACH 8 VARIABLE DENSITY	3234		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER MODEL — AUG 01,1984

MODEL ID : 4

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
PRE-ATP/001	.0140	OH1A-2	9*19*72	09*26*72	10/100	120	LARC MACH 8 VARIABLE DENSITY	3234		UNASSIGNED
PRE-ATP/001	.0140	OH1B	11* 6*72	11*08*72	40/ 24	35	LARC MACH 8 VARIABLE DENSITY	3283		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 5

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
PRE-ATP/001	.0058	OH1A-3	9*19*72 - 09*26*72	10/100	120	LARC MACH 8 VARIABLE DENSITY	3234		UNASSIGNED

MODEL ID : 6

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
ATP	.0150	OA4	10* 2*72 - 10*17*72	200/176	54	ARC 3.5-FT HYPERSONIC	147	2007	PUBLISHED
ATP	.0150	OA3	10*24*72 - 11*10*72	200/320	214	ARC 6X6-FT SUPERSONIC	650	2009	PUBLISHED
ATP	.0150	LA1	11*19*72 - 12*19*72	60/ 84	73	LARC 8-FT TRANSONIC PRESSURE	626	2002	PUBLISHED
ATP	.0150	OA7	11*27*72 - 12*08*72	100/100	110	LARC UNITARY PLAN	1007	2014	PUBLISHED
ATP	.0150	IA8	2*12*73 - 03*12*73	80/160	54	ARC 14-FT TRANSONIC	711	2173	PUBLISHED
2A/089B	.0150	MA7	5*14*73 - 05*18*73	50/ 50	81	LARC UNITARY PLAN	1031	2069	PUBLISHED
2A/089B	.0150	MA28	9*29*76 - 09*29*76	7/ 1	0	AEDC A / SUPERSONIC	K8A		UNASSIGNED

C-4

MODEL ID : 7

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
PRE-ATP/001	.0190	IA2	10*11*72 -	11*03*72	40/244	92	ARC 9X7-FT SUPERSONIC	616	2013	PUBLISHED
PRE-ATP/001	.0190	IA7	2*12*73 -	02*23*73	80/160	85	ARC 11-FT TRANSONIC	686	2024	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 8

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
140A,B/747	.0460	CS1	6* 9*75 - 06*13*75	80/ 95	165	UNIV. OF WASH. LOW SPEED	1160		UNASSIGNED
140A,B/747	.0460	CS3	9*12*75 - 09*15*75	40/ 80	129	UNIV. OF WASH. LOW SPEED	1170	2338	PUBLISHED
140A,B/747	.0460	CS6	2* 5*76 - 02*11*76	60/ 58	203	GENERAL DYNAMIC - LOW SPEED	691		UNASSIGNED

MODEL ID : 9

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
PRE-ATP/001	.0075	IA4	11* 2*72 - 11*17*72	80/ 75	62	LTV 4X4-FT SUPERSONIC	458	2015	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 10

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
PRE-ATP/001	.0193	MA5	9*15*72 -	09*25*72	80/ 60	30	LARC UNITARY PLAN	1002	2001	PUBLISHED
PRE-ATP/001	.0193	FA1	10*10*72 -	11*15*72	416/400	200	LARC 16-FT TRANSONIC DYNAMIC	210		UNASSIGNED
PRE-ATP/001	.0193	IA3	11* 3*72 -	11*16*72	24/ 41	53	RI 7X11-FT LOW SPEED	693	2018	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 11

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B	.0400	IS1B	7*23*73 - 08*01*73	60/ 64	4	ARC 9X7-FT SUPERSONIC	705	2401	PUBLISHED
2A/089B	.0400	IS1A	8* 2*73 - 08*08*73	60/ 60	21	ARC 11-FT TRANSONIC	705	2401	PUBLISHED
2A/089B	.0400	IS1C	8* 9*73 - 08*11*73	12/ 24	3	ARC 8X7-FT SUPERSONIC	705	2401	PUBLISHED
2A/089B	.0400	OS3	8*11*73 - 08*14*73	58/ 52	31	ARC 8X7-FT SUPERSONIC	705	2401	PUBLISHED

MODEL #12 WAS NOT BUILT.....

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 13

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B	.0040	IS6B	3*20*73 -	05*27*73	50/ 50	70	MSFC 14-IN TRANSONIC	559	2158	PUBLISHED
2A/089B	.0040	OA47	3*28*73 -	04*05*73	116/116	245	MSFC 14-IN TRANSONIC	568	2029	PUBLISHED
2A/089B	.0040	IA31FA	4* 9*73 -	04*13*73	60/ 60	104	MSFC 14-IN TRANSONIC	566	2026	PUBLISHED
2A/089B	.0040	IA31FB	4*13*73 -	04*30*73	50/271	220	MSFC 14-IN TRANSONIC	570	2028	PUBLISHED
2A/089B	.0040	IA6	4*30*73 -	05*03*73	45/ 52	94	MSFC 14-IN TRANSONIC	571	2039	PUBLISHED
2A/089B	.0040	IA32F	5* 9*73 -	05*24*73	100/180	190	MSFC 14-IN TRANSONIC	567	2027	PUBLISHED
2A/089B	.0040	IA31FC	6*21*73 -	07*09*73	32/ 51	145	MSFC 14-IN TRANSONIC	573	2072	PUBLISHED
2A/089B	.0040	IS6A	10* 2*73 -	10*11*73	80/ 74	126	MSFC 14-IN TRANSONIC	582	2158	PUBLISHED
2A/089B	.0040	IA53	12*20*73 -	01*04*74	40/ 36	45	MSFC 14-IN TRANSONIC	588	2123	PUBLISHED
2A/089B	.0040	IA68	1*18*74 -	01*29*74	32/ 36	34	RI 7-FT TRISONIC	281	2144	PUBLISHED
089B	.0040	LA88	5*21*75 -	05*21*75	16/ 16	6	LARC 20-IN HYPERSONIC (M=6)	6468	2311	PUBLISHED
089B	.0040	LA87	8*26*75 -	08*29*75	36/ 36	4	LARC 4-FT HYPERSONIC	446	2311	PUBLISHED
089B	.0040	LA78	1*15*76 -	01*28*76	16/ 16	4	LARC 4-FT HYPERSONIC	267	2311	PUBLISHED
140C	.0040	LA85	4* 7*76 -	05*24*76	88/ 88	64	LARC 22-IN HELIUM	445	2343	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 14

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B(MOD)	.0190	IA12B	4*23*73 - 05*07*73	120/156	63	ARC 9X7-FT SUPERSONIC	710	2048	PUBLISHED
2A/089(MOD)	.0190	IA36	6*15*73 - 06*22*73	60/ 80	120	CALSPAN 8-FT TRANSONIC	053	2064	PUBLISHED
2A/089(MOD)	.0190	IA12C	7*11*73 - 07*27*73	220/220	133	ARC 8X7-FT SUPERSONIC	710	2065	PUBLISHED

MODEL ID : 15

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
TPS TILES	1.0000	FH1	11*15*72 - 01*01*73	160/ 80	200	LARC HIGH RE'S NUMBER HELIUM	100		UNASSIGNED
TPS TILES	1.0000	OH2	4*18*73 - 06*01*73	40/144	81	ARC 3.5-FT HYPERSONIC	158	2035	PUBLISHED
TPS TILES	1.0000	OH43	12* 2*73 - 12*21*73	160/128	92	ARC 3.5-FT HYPERSONIC	182	2250	PUBLISHED
TPS TILES	1.0000	IH27	9* 7*74 - 09*25*74	80/196	65	ARC 3.5-FT HYPERSONIC	200	2210	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 16

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140A,B	.0405	OA110	3*15*74 - 03*20*74	80/ 48	85	RI 7X11-FT LOW SPEED	721	2155	PUBLISHED
4/140A,B	.0405	OA119A	6*17*74 - 06*25*74	20/ 45	45	RI 7X11-FT LOW SPEED	726	2187	PUBLISHED
4/140A,B	.0405	OA119B	8*22*74 - 09*06*74	60/100	213	RI 7X11-FT LOW SPEED	730	2203	PUBLISHED
4/140A,B	.0405	OA143	11* 6*74 - 11*11*74	40/ 55	60	RI 7X11-FT LOW SPEED	737	2221	PUBLISHED
4/140A,B	.0405	OA163A	11*24*75 - 12*09*75	160/144	215	RI 7X11-FT LOW SPEED	751	2289	PUBLISHED
4/140A,B	.0405	OA163B	12*21*76 - 12*23*76	35/ 35	99	RI 7X11-FT LOW SPEED	788	2361	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 17

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B	.0300	IA9A	4* 2*73 - 04*14*73	90/113	118	ARC 11-FT TRANSONIC	707	2032	PUBLISHED
2A/089B	.0300	OA12A	4*12*73 - 04*23*73	90/103	98	ARC 11-FT TRANSONIC	707	2032	PUBLISHED
2A/089B	.0300	IA9C	4*22*73 - 05*01*73	60/ 60	102	ARC 8X7-FT SUPERSONIC	707	2032	PUBLISHED
2A/089B	.0300	IA9B	5* 2*73 - 05*09*73	100/120	65	ARC 9X7-FT SUPERSONIC	707	2032	PUBLISHED
2A/089B	.0300	OA12C	5* 2*73 - 05*10*73	60/ 60	46	ARC 8X7-FT SUPERSONIC	707	2032	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01.1984

MODEL ID : 18

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/O89B	.0150	OA11A	4* 9*73 - 04*17*73	144/176	62	ARC 3.5-FT HYPERSONIC	157	2044	PUBLISHED
2A/O89B	.0150	OA43	4*18*73 - 05*04*73	128/160	137	ARC 6X6-FT SUPERSONIC	706	2050	PUBLISHED
2A/O89B	.0150	OA11B	5*14*73 - 05*25*73	140/160	70	ARC 3.5-FT HYPERSONIC	160	2059	PUBLISHED
2A/O89B	.0150	OA44-1	6* 1*73 - 06*08*73	40/ 54	47	LARC UNITARY PLAN	1035	2057	PUBLISHED
2A/O89B	.0150	OA17-2	6*18*73 - 07*06*73	20/100	55	LARC LOW TURBULANCE PRESSURE	138	2058	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 19

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL	NO.	DOCUMENT NO.	STATUS
2A/089B		.0225	IH5	1*21*74 -	07*22*74	120/105	106	CALSPAN 32-IN	LUDWIEG	181	2308	PUBLISHED
5/140C		.0225	IH34	5* 5*75 -	09*03*75	240/264	57	LERC 10X10-FT	SUPERSONIC	038	2282	PUBLISHED
5/140C		.0225	IH39	9*22*76 -	04*14*77	240/226	163	LERC 10X10-FT	SUPERSONIC	041	2435	PUBLISHED
5/140C		.0225	IH75	10* 3*77 -	12*12*77	200/320	41	CALSPAN 32-IN	LUDWIEG	100	2453	PUBLISHED
5/140C		.0225	IH83	1*25*78 -	03*10*78	200/102	41	LERC 10X10-FT	SUPERSONIC	044	2440	PUBLISHED

MODEL # 20 WAS NOT BUILT

MODEL ID : 21

REF.	MODEL SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139B	.0175	OH3A	6*28*73 - 06*30*73	40/ 16	36	AEDC B / HYPERSONIC	288	2100	PUBLISHED
3/139B	.0175	OH3B	7* 9*73 - 07*11*73	40/ 23	147	AEDC B / HYPERSONIC	289	2100	PUBLISHED
3/139B	.0175	OH4C	9*26*73 - 09*26*73	8/ 8	60	AEDC B / HYPERSONIC	352	2225	PUBLISHED
3/139B	.0175	OA99	3*26*74 - 4*12*74	50/ 52	14	LARC 60-FT. VACUUM SPHERE	3289	2172	PUBLISHED
3/139B	.0175	OH25A	8*21*74 - 08*22*74	12/ 12	82	AEDC B / HYPERSONIC	83A	2252	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 22

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139	.0175	OH4B	9*29*73 - 10*04*73	48/ 38	224	AEDC B / HYPERSONIC	352	2099	PUBLISHED
3/139	.0175	IH3	10*31*73 - 11*09*73	128/128	79	ARC 3.5-FT HYPERSONIC	178	2136	PUBLISHED
3/139	.0175	IH1	12* 3*73 - 12*14*73	100/100	44	LARC UNITARY PLAN	1071	2153	PUBLISHED
3/139	.0175	IH20	1*18*74 - 02*06*74	192/192	105	ARC 3.5-FT HYPERSONIC	185	2148	PUBLISHED
3/139	.0175	FH10	1*21*74 - 01*29*74	32/ 32	9	AEDC F / HYPERSONIC	291	2197	PUBLISHED
3/139	.0175	OH6	2* 6*74 - 02*11*74	148/ 56	39	ARC 3.5-FT HYPERSONIC	183	2151	PUBLISHED
3/139B	.0175	OH49A	4* 3*74 - 04*06*74	216/ 17	87	AEDC B / HYPERSONIC	525	2355	PUBLISHED
4/140B	.0175	OH49B	7* 2*74 - 07*12*74	72/ 67	454	AEDC B / HYPERSONIC	57A	2222	PUBLISHED
4/140B	.0175	OH26	7*22*74 - 07*29*74	80/ 96	56	ARC 3.5-FT HYPERSONIC	199	2193	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL   AUG 01,1984

MODEL ID : 23

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL	NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B	.0200	OS1	8* 6*73	- 08*10*73	80/ 72	39	LARC 26-IN	TRANSONIC BLOWDOWN	545	2094	PUBLISHED

MODEL ID : 24

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
ATP		.0250	OS2	6* 4*73	06*07*73	120/ 24	18	LARC 26-IN TRANSONIC BLOWDOWN	544	2067	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 25

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B	.0400	IA109	7*26*70 - 8* 8*74	40/100	19	MSFC IMPULSE BASE FLOW FAC.	27	2382	PUBLISHED
2A/089B	.0400	OH8F	5*15*74 - 07*16*74	340/334	66	MSFC IMPULSE BASE FLOW FAC.	027	2382	PUBLISHED
2A/089B	.0400	OH64	4*14*75 - 06*20*75	200/450	200	LERC SPACE POWER FACILITY	OH64	2288	PUBLISHED

AUG 01, 1984

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 26

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	STATUS
3/139		.0100	OH10	8*17*73 - 09*04*73	96/104	35	ARC 3.5-FT HYPERSONIC	171	2085	PUBLISHED
3/139		.0100	IH2	9* 4*73 - 09*11*73	80/104	21	ARC 3.5-FT HYPERSONIC	171	2085	PUBLISHED
3/139		.0100	IH4	11*12*73 - 11*16*73	40/ 64	47	LARC UNITARY PLAN	1059	2138	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 27

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
RI PRR ORB.	.0150	MA6	4* 2*73 -	04*06*73	120/136	4	ARC 3.5-FT HYPERSONIC	156		UNASSIGNED

MODEL # 28 WAS NOT BUILT

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 29

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL	NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139	.0175	OH9	9*13*73	- 09*21*73	16/ 16	61	AEDC B /	HYPERSONIC	353	2251	PUBLISHED
3/139	.0175	OH11	10*24*73	- 11*01*73	40/ 37	23	AEDC F /	HYPERSONIC	VA35	2141	PUBLISHED
3/139	.0175	OH4A	11*12*73	- 12*05*73	20/ 20	57	AEDC B /	HYPERSONIC	352	2154	PUBLISHED
3/139B	.0175	OH52	5* 6*74	- 05*15*74	16/ 16	32	AEDC B /	HYPERSONIC	524	2330	PUBLISHED
4/140B	.0175	MH2	9* 3*75	- 01*23*76	16/ 11	22	AEDC B /	HYPERSONIC	D5A		UNASSIGNED
4/140B	.0175	MH1	1*13*76	- 01*23*76	24/ 64	30	AEDC F /	HYPERSONIC	29A		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER MODEL　　AUG 01,1984

MODEL ID : 30

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B	.0125	IS4	10*18*73 -	10*24*73	120/ 58	94	LARC 26-IN TRANSONIC BLOWDOWN	547	2146	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 31

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B		.0060	OH40	1*30*73	02*05*73	36/ 40	52	LARC MACH 8 VARIABLE DENSITY	3619	2049	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 32

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
3/139B	.0100	IA13	7* 5*73 - 07*17*73	40/ 39	762	AEDC A / SUPERSONIC	323	2062	PUBLISHED
3/139B	.0100	OA23-2	7*26*73 - 07*31*73	80/ 90	39	ARC 3.5-FT HYPERSONIC	168	2071	PUBLISHED
3/139B	.0100	IA10	8* 1*73 - 08*03*73	50/ 40	18	ARC 3.5-FT HYPERSONIC	169	2078	PUBLISHED
3/139B	.0100	LA25	8*30*73 - 09*07*73	40/ 48	126	LARC 31-IN CONT-FLOW HYP.	100	2126	CANCEL
3/139B	.0100	IA15	10*10*73 - 10*16*73	64/ 80	25	ARC 3.5-FT HYPERSONIC	175	2102	PUBLISHED
3/139B	.0100	OA85	10*31*73 - 11*08*73	50/ 60	75	LARC 31-IN CONT-FLOW HYP.	101	2113	PUBLISHED
3/139B	.0100	LA35	11*12*73 - 11*13*73	16/ 20	19	LARC 31-IN CONT-FLOW HYP.	102	2127	PUBLISHED
3/139,089B	.0100	IA57	11*20*73 - 11*20*73	10/ 9	10	AEDC A / SUPERSONIC	422	2112	PUBLISHED
3/139,089B	.0100	IA61A	1*30*74 - 01*31*74	10/ 10	88	AEDC A / SUPERSONIC	422	2143	PUBLISHED
3/139,089B	.0100	IA58	2*11*74 - 02*13*74	32/ 40	34	LARC 31-IN CONT-FLOW HYP.	107	2133	PUBLISHED
3/139,089B	.0100	IA60	2*14*74 - 02*20*74	15/ 36	55	LARC 31-IN CONT-FLOW HYP.	108	2137	PUBLISHED
4/140A,B	.0100	OA105	2*20*74 - 2*22*74	16/ 20	50	LARC 31-IN CONT-FLOW HYP.	109	2137	PUBLISHED
4/140A,B	.0100	OA82	8*12*74 - 08*16*74	40/ 48	96	LARC 31-IN CONT-FLOW HYP.	113	2195	PUBLISHED
~/140A,B	.0100	MA22	5* 6*75 - 06*03*75	100/ 168	357	LARC 31-IN CONT-FLOW HYP.	118	2267	PUBLISHED
140A,B	.0100	LA36B	6* 3*75 - 06*05*75	75/ 27	41	LARC LOW TURBULANCE PRESSURE	214	2292	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 33

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B	.0060	OH41A	3*19*73 - 03*28*73	40/ 64	78	LARC MACH 8 VARIABLE DENSITY	3778	2075	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 34

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139B,W/CANS	.0040	OA48	5*25*73	6*11*73	100/166	364	MSFC 14-IN TRANSONIC	574	2055	PUBLISHED
3A/139B	.0040	IA37A	7*10*73	07*13*73	60/ 36	64	MSFC 14-IN TRANSONIC	579	2063	PUBLISHED
3A/139B	.0040	IA48	7*18*73	07*21*73	20/ 24	40	MSFC 14-IN TRANSONIC	580	2063	PUBLISHED
3A/139B	.0040	OA72	7*30*73	08*24*73	40/176	42	LARC 22-IN HELIUM	415	2092	PUBLISHED
3A/139B	.0040	IA52	10*11*73	10*17*73	16/ 28	27	MSFC 14-IN TRANSONIC	584	2042	PUBLISHED
3A/139B	.0040	IA37B	10*15*73	10*16*73	16/ 22	42	MSFC 14-IN TRANSONIC	585	2093	PUBLISHED
4/140A,B	.0040	OA49	10*18*73	11*09*73	198/170	415	MSFC 14-IN TRANSONIC	581	2095	PUBLISHED
4/140A,B	.0040	IA62F	11*15*73	11*19*73	16/ 19	33	MSFC 14-IN TRANSONIC	589	2103	PUBLISHED
4/140A,B	.0040	OA88	12*11*73	12*28*73	60/ 60	191	LARC 22-IN HELIUM	7422	2125	PUBLISHED
5/140C	.0040	MA21	8*15*75	09*04*75	80/ 92	50	JPL 20-IN SUPERSONIC	702		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 35

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
TILE PANEL	.1820	OS32	7•15•76 - 07•27•76	96/ 80	89	ARC 2X2-FT TRANSONIC	167	2339	IN PROCESS

MODEL ID : 36

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140A,B	.0150	IA29	9*12*73 -	09*25*73	80/184	111	ARC 6X6-FT SUPERSONIC	630	2077	PUBLISHED
4/140A,B	.0150	0A63	9*25*73 -	9*28*73	64/ 80	98	ARC 6X6-FT SUPERSONIC	630	2077	PUBLISHED
4/140A,B	.0150	0A64	10*30*73 -	10*31*73	50/ 30	28	LARC UNITARY PLAN	1063	2108	PUBLISHED
4/140A,B	.0150	IA35	11* 1*73 -	11*02*73	60/ 30	22	LARC UNITARY PLAN	1063	2108	PUBLISHED
4/140A,B	.0150	IA16	11*17*73 -	12*04*73	80/ 52	9	ARC 3.5-FT HYPERSONIC	180	2124	PUBLISHED
4/140A,B	.0150	0A26	11*17*73 -	12*04*73	64/140	27	ARC 3.5-FT HYPERSONIC	180	2124	PUBLISHED
4/140A,B	.0150	0A83	5* 8*74 -	05*16*74	80/160	34	ARC 3.5-FT HYPERSONIC	194	2177	PUBLISHED
4/140A,B	.0150	0A102	6*17*74 -	06*18*74	18/.18	10	LARC 8-FT TRANSONIC PRESSURE	687	2229	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 37

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139	.0100	IH21	10*29*73 -	12*13*73	80/145	31	CALSPAN HYPERSONIC SHOCK	100	2164	PUBLISHED
3/139	.0100	OH12	10*29*73 -	12*13*73	80/145	32	CALSPAN HYPERSONIC SHOCK	100	2164	PUBLISHED
5/140C	.0100	IH33A	10*14*74 -	10*18*74	32/ 32	10	CALSPAN HYPERSONIC SHOCK	120	2249	PUBLISHED
5/140C	.0100	IH33B	12* 5*74 -	12*19*74	48/ 80	24	CALSPAN HYPERSONIC SHOCK	131	2249	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 38

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
PRE-ATP/001	.0058	OH1A-4	9*19*72	09*26*72	10/100	120	LARC MACH 8 VARIABLE DENSITY	3234		UNASSIGNED
2A/089B	.0058	OH41B	5* 8*73	05*10*73	40/ 24	20	LARC MACH 8 VARIABLE DENSITY	4060	2076	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 39

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
VEH 102	.0500	OA223	11*20*76 - 11*30*76	40/ 88	13	RI 7X11-FT LOW SPEED	776	2402	PUBLISHED
VEH 102	.0500	OA145A	3* 8*77 - 04*02*77	160/480	981	ARC 11-FT TRANSONIC	118	2380	PUBLISHED
VEH 102	.0500	OA145C	4* 6*77 - 04*20*77	80/100	188	ARC 8X7-FT SUPERSONIC	118	2389	PUBLISHED
VEH 102	.0500	OA145B	4*15*77 - 05*03*77	80/348	240	ARC 9X7-FT SUPERSONIC	118	2364	PUBLISHED
VEH 102	.0500	OA101	9*13*77 - 11*11*77	160/160	373	ARC 12-FT PRESSURE	218	2405	PUBLISHED
VEH 102	.0500	OA270A	5*15*78 - 06*09*78	60/160	156	LARC 16-FT TRANSONIC	325	2430	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 40

REF.	MODEL SCALE	TEST NO.	TESTING SCHED. - COMPL.	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
PANELS	1.0000	OS4A	9* 9*74 - 10*09*74	154/260	36	ARC 2X2-FT TRANSONIC	041	2450	PUBLISHED
PANELS	1.0000	OS4B	11*18*75 - 01*15*76	140/ 96	0	ARC 2X2-FT TRANSONIC	154	2450	PUBLISHED

MODEL ID : 41

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
2A/089B	.0060	OH13	6*13*73	- 06*13*73	8/ 8	18	LARC MACH 8 VARIABLE DENSITY	644	2096	PUBLISHED
2A/089B	.0060	IH16	7* 6*73	- 07*13*73	35/ 80	12	LARC UNITARY PLAN	1041	2166	PUBLISHED
2A/089B	.0060	IH15	8*13*73	- 08*17*73	64/ 72	30	ARC 3.5-FT HYPERSONIC	172	2098	PUBLISHED
2A/089B	.0060	IH17	10* 9*73	- 10*16*73	40/ 48	59	LARC MACH 8 VARIABLE DENSITY	646	2105	PUBLISHED
2A/089B	.0060	IH18	10*19*73	- 10*30*73	40/ 40	22	LARC 20-IN FREON	118	2110	PUBLISHED

MODEL ID : 42

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139B	.0150	0A58	6* 4*73 -	06*18*73	80/ 76	38	ARC 3.5-FT HYPERSONIC	163	2060	PUBLISHED
3/139B	.0150	0A44-2	6*11*73 -	06*15*73	40/ 54	36	LARC UNITARY PLAN	1035	2057	PUBLISHED
3/139B	.0150	0A17-1	6*18*73 -	07*06*73	60/124	65	LARC LOW TURBULANCE PRESSURE	138	2058	PUBLISHED
3/139B	.0150	0A73	7*11*73 -	7*18*73	60/ 96	37	ARC 3.5-FT HYPERSONIC	167	2082	PUBLISHED
3/139B	.0150	0A70	7*20*73 -	7*26*73	30/ 40	66	LARC UNITARY PLAN	1043	2073	PUBLISHED
140A,B	.0150	LA36A	11* 5*73 -	11*11*73	75/ 75	22	LARC 16-FT TRANSONIC	149		UNASSIGNED
140A,B	.0150	LA58	9*30*74 -	10*04*74	49/ 80	72	LTV 4X4-FT SUPERSONIC	512	2215	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 43

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139B	.0405	OA18	5* 8*73 - 05*17*73	100/114	189	RI 7X11-FT LOW SPEED	704	2045	PUBLISHED
3/139B	.0405	OA21A	5*21*73 - 06*04*73	100/ 72	348	RI 7X11-FT LOW SPEED	705	2053	PUBLISHED
3/139B W/CANS	.0405	OA21B	6*21*73 - 06*25*73	40/ 55	99	RI 7X11-FT LOW SPEED	705	2053	PUBLISHED
3/139B	.0405	OA69	8*28*73 - 09*01*73	80/ 71	205	RI 7X11-FT LOW SPEED	711	2081	PUBLISHED
3/139B	.0405	OA71C	9* 4*73 - 09*14*73	100/139	71	RI 7X11-FT LOW SPEED	712	2086	PUBLISHED
4/140A.B	.0405	OA62A	10* 5*73 - 10*23*73	120/195	98	RI 7X11-FT LOW SPEED	715	2097	PUBLISHED
4/140A.B	.0405	OA86	10*26*73 - 11*09*73	80/174	331	RI 7X11-FT LOW SPEED	716	2114	PUBLISHED
4/140A.B	.0405	OA62B	11*13*73 - 12*06*73	100/240	448	RI 7X11-FT LOW SPEED	717	2104	PUBLISHED
4/140A.B	.0405	OA118	4*24*74 - 04*26*74	48/ 40	54	RI 7X11-FT LOW SPEED	724	2139	PUBLISHED
4/140A.B/747	.0405	CA4	5*28*74 - 06*07*74	64/120	100	UNIV. OF WASH. LOW SPEED	1128		UNASSIGNED
4/140A.B/C-5A	.0405	CA2-1	6* 4*74 - 06*10*74	120/ 80	100	LOCKHEED (GA) - LOW SPEED			UNASSIGNED
4/140A.B/747	.0405	CA3	8*15*74 - 08*30*74	120/131	194	UNIV. OF WASH. LOW SPEED	1136	2201	PUBLISHED
4/140A.B (ALT)	.0405	OA123	9* 6*74 - 09*10*74	40/ 47	41	RI 7X11-FT LOW SPEED	731	2202	PUBLISHED
4/140A.B	.0405	OA124	10*14*74 - 10*23*74	60/ 60	127	RI 7X11-FT LOW SPEED	736	2209	PUBLISHED
4/140A.B/747	.0405	CA8	8*18*75 - 09*12*75	200/324	536	LARC V/STOL	129	2290	PUBLISHED
4/140A.B/747	.0405	CA15A	10*16*75 - 11*01*75	240/239	379	UNIV. OF WASH. LOW SPEED	1173	2347	PUBLISHED
4/140A.B/747	.0405	CA15B	11*19*75 - 11*26*75	75/110	93	UNIV. OF WASH. LOW SPEED	1178	2348	PUBLISHED
4/140A.B(ALT)	.0405	OA172	12*15*75 - 01*13*76	120/210	122	RI 7X11-FT LOW SPEED	752	2294	PUBLISHED
4/140A.B(ALT)	.0405	OA176	3*29*76 - 04*15*76	60/ 83	113	RI 7X11-FT LOW SPEED	754	2314	PUBLISHED
4/140A.B/747	.0405	CA17	6*21*76 - 07*02*76	152/152	261	UNIV. OF WASH. LOW SPEED	1184	2349	PUBLISHED

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MODEL ID : 44

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
140C/REMOTE ELE	.0150	LA62	5*14*75 -	05*23*75	40/ 80	301	LARC 8-FT TRANSONIC PRESSURE	717	2264	PUBLISHED
140C/REMOTE ELE	.0150	LA67	6*20*75 -	07*02*75	40/120	131	LTV 4X4-FT SUPERSONIC	552	2266	PUBLISHED
140C/REMOTE ELE	.0150	LA63A	7*18*75 -	07*18*75	40/ 14	63	LARC UNITARY PLAN	1118	2270	PUBLISHED
140C/REMOTE ELE	.0150	LA70	7*28*75 -	08*06*75	38/ 60	299	CALSPAN 8-FT TRANSONIC	103	2269	PUBLISHED
140C/REMOTE ELE	.0150	LA61A	8*25*75 -	09*10*75	40/ 40	138	LARC LOW TURBULANCE PRESSURE	219	2278	CANCEL
140C/REMOTE ELE	.0150	LA63B	9*12*75 -	09*17*75	40/ 38	191	LARC UNITARY PLAN	1151	2279	PUBLISHED
140C/REMOTE ELE	.0150	LA66	10*20*75 -	10*24*75	80/ 80	26	ARC 12-FT PRESSURE	135	2281	PUBLISHED
140C/REMOTE ELE	.0150	LA61B	1* 5*76 -	01*14*76	40/ 96	81	LARC LOW TURBULANCE PRESSURE	228	2300	PUBLISHED
140C/REMOTE ELE	.0150	LA75	2*25*76 -	03*06*76	48/128	141	LTV 4X4-FT SUPERSONIC	573	2305	PUBLISHED
140C/REMOTE ELE	.0150	LA[illegible]5	4* 6*76 -	04*16*76	90/ 90	283	LARC UNITARY PLAN	1173	2318	PUBLISHED
140C/REMOTE ELE	.0150	LA77	7* 9*76 -	07*24*76	120/151	521	ARC 11-FT TRANSONIC	200	2344	PUBLISHED
140C/REMOTE ELE	.0150	LA91	9* 3*76 -	09*15*76	80/104	214	LARC 8-FT TRANSONIC PRESSURE	758	2352	PUBLISHED
140C/REMOTE ELE	.0150	LA101	5*18*77 -	05*24*77	55/ 55	200	LARC UNITARY PLAN	1194	2390	PUBLISHED
140C SILTS	.0150	LA111	8* 3*77 -	08*05*77	95/ 40	95	LARC 8-FT TRANSONIC PRESSURE	786	2395	PUBLISHED
140C SILTS	.0150	LA110	8* 8*77 -	08*10*77	30/ 30	60	LARC UNITARY PLAN	1212	2396	PUBLISHED
140C SILTS	.0150	LA114	8*23*77 -	08*31*77	30/ 60	70	LARC UNITARY PLAN	1217	2399	PUBLISHED
140C/REMOTE ELE	.0150	LA115	2* 1*78 -	02*06*78	45/ 45	175	LARC 8-FT TRANSONIC PRESSURE	803	2409	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 45

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
140A,B/747	.0300	CA5	9*20*74 - 09*30*74	144/181	520	THE BOEING CO. - TRANSONIC	1431	2211	PUBLISHED
140A,B/747	.0300	CA20	10* 9*74 - 10*15*74	115/115	288	THE BOEING CO. - TRANSONIC	1431	2217	PUBLISHED
140A,B (MOD)	.0300	OA161A	3*10*75 - 03*20*75	140/160	285	ARC 11-FT TRANSONIC	094	2245	PUBLISHED
140A,B (MOD)	.0300	OA161B	3*20*75 - 03*26*75	24/ 30	49	ARC 9X7-FT SUPERSONIC	094	2245	PUBLISHED
140A,B (MOD)	.0300	OA161C	3*26*75 - 03*31*75	20/ 22	45	ARC 8X7-FT SUPERSONIC	094	2245	PUBLISHED
140A,B/747	.0300	CA6	5*20*75 - 06*06*75	200/265	509	THE BOEING CO. - TRANSONIC	1472	2262	PUBLISHED
140A,B/747	.0300	CS2	6* 9*75 - 06*16*75	95/ 95	165	THE BOEING CO. - TRANSONIC	1474		UNASSIGNED
140A,B/(ALT)	.0300	OA159	6*23*75 - 07*08*75	160/152	50	ARC 12-FT PRESSURE	078	2265	PUBLISHED
140A,B/747	.0300	CA16	8*23*75 - 09*05*75	72/ 84	60	TEXAS A+M 7X10-FT LOW SPEED	7515		UNASSIGNED
140A,B/747	.0300	CS4	9*29*75 - 10*02*75	40/ 64	95	THE BOEING CO. - TRANSONIC	1490	2341	PUBLISHED
140A,B/747	.0300	CS5	11* 3*75 - 11*05*75	24/ 33	192	THE BOEING CO. - TRANSONIC	1493	2341	PUBLISHED
140A,B/747	.0300	CA14	11*13*75 - 12*02*75	160/236	850	THE BOEING CO. - TRANSONIC	1496	2307	PUBLISHED
140C(ALT)	.0300	OA173	3*15*76 - 03*26*76	160/256	48	ARC 12-FT PRESSURE	180	2304	PUBLISHED
140C(ALT)/747	.0300	CA13	6* 8*76 - 07*01*76	160/193	54	ARC 14-FT TRANSONIC	121	2332	PUBLISHED
140C(ALT)	.0300	OA250	7* 1*77 - 07*07*77	32/ 34	23	RI 7X11-FT LOW SPEED	775	2392	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01.1984

MODEL ID : 46

REF. MODEL	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139, 139A	.0060	OH42A	5*14*73 - 05*16*73	20/ 20	20	LARC MACH 8 VARIABLE DENSITY	4080	2101	PUBLISHED
3/139, 139A	.0060	OH42B	5*25*73 - 06*01*73	40/ 48	64	LARC MACH 8 VARIABLE DENSITY	4080	2101	PUBLISHED
3/139A, W/CAN	.0060	OH42C	6*14*73 - 06*15*73	20/ 16	26	LARC MACH 8 VARIABLE DENSITY	4080	2101	PUBLISHED
3/139B	.0060	OH51-2	6*26*74 - 07*03*74	12/180	280	LARC 31-IN CONT-FLOW HYP.	112	2368	PUBLISHED

MODEL ID : 47

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140A,B	.0300	IA14A	9* 4*73 - 09*13*73	130/151	149	ARC 11-FT TRANSONIC	716	2084	PUBLISHED
4/140A,B	.0300	OA22A	9*12*73 - 09*14*73	20/ 21	24	ARC 11-FT TRANSONIC	716	2130	PUBLISHED
4/140A,B	.0300	IA14B	9*14*73 - 09*19*73	48/ 41	66	ARC 9X7-FT SUPERSONIC	716	2129	PUBLISHED
4/140A,B	.0300	OA22B	9*19*73 - 09*20*73	40/ 31	30	ARC 9X7-FT SUPERSONIC	716	2131	PUBLISHED
4/140A,B	.0300	OA53B	11*12*73 - 11*16*73	60/160	103	ARC 9X7-FT SUPERSONIC	747	2178	PUBLISHED
4/140A,B	.0300	OA53A	11*19*73 - 11*27*73	128/128	267	ARC 11-FT TRANSONIC	747	2128	PUBLISHED
4/140A,B	.0300	OA53C	11*28*73 - 12*06*73	60/159	159	ARC 8X7-FT SUPERSONIC	747	2185	PUBLISHED
4/140A,B	.0300	OA37	1* 7*74 - 01*25*74	80/103	112	RI 7X11-FT LOW SPEED	719	2140	PUBLISHED
4/140A,B (MOD)	.0300	IA81A	7*26*74 - 08*27*74	84/184	99	ARC 11-FT TRANSONIC	019	2169	PUBLISHED
4/140A,B (MOD)	.0300	IA81B	8* 9*74 - 08*22*74	60/208	88	ARC 9X7-FT SUPERSONIC	019	2194	PUBLISHED
4/140A,B (MOD)	.0300	OA155	2*10*75 - 03*07*75	80/152	205	LARC V/STOL	114	2237	IN PROCESS
4/140A,B (MOD)	.0300	CA148	5* 5*75 - 05*17*75	220/264	474	ARC 11-FT TRANSONIC	073	2254	PUBLISHED
4/140A,B/747	.0300	CA9	6*25*75 - 07*14*75	320/302	85	THE BOEING CO. - TRANSONIC	1477	2268	PUBLISHED
4/140A,B (MOD)	.0300	IA135A	3* 2*76 - 03*23*76	120/146	132	ARC 11-FT TRANSONIC	144	2306	PUBLISHED
4/140A,B (MOD)	.0300	IA135B	3* 5*76 - 03*23*76	60/100	50	ARC 9X7-FT SUPERSONIC	144	2306	PUBLISHED
4/140A,B (MOD)	.0300	IA135C	3*12*76 - 03*23*76	20/ 40	5	ARC 8X7-FT SUPERSONIC	144	2306	PUBLISHED
140A,B (ALT)	.0300	OA175	6*28*76 - 07*09*76	160/240	290	ARC 11-FT TRANSONIC	187	2333	PUBLISHED
5/140C	.0300	OA149B	2* 2*77 - 02*07*77	40/168	201	ARC 9X7-FT SUPERSONIC	115	2370	PUBLISHED
5/140C	.0300	OA149C	2*16*77 - 02*18*77	40/144	25	ARC 8X7-FT SUPERSONIC	115	2370	PUBLISHED
5/140C	.0300	OA149A	2*24*77 - 03*04*77	160/144	390	ARC 11-FT TRANSONIC	115	2376	PUBLISHED
5/140C	.0300	IA105A	9* 2*77 - 11*20*77	290/281	885	AEDC 16-FT TRANSONIC	470	2398	PUBLISHED
5/140C	.0300	IA105B	1* 9*78 - 02*01*78	100/258	143	ARC 9X7-FT SUPERSONIC	242	2413	PUBLISHED
5/140C	.0300	OA126B	4*17*78 - 04*30*78	120/ 97	256	ARC 9X7-FT SUPERSONIC	289	2424	PUBLISHED
5/140C	.0300	OA126A	5* 1*78 - 05*30*78	240/131	304	ARC 11-FT TRANSONIC	289	2424	PUBLISHED

VEH102	.0300	OA129	7* 7*78 - 07*15*78	40/ 64	477	AEDC 16-FT TRANSONIC	507	2434	PUBLISHED
5/140C	.0300	IA182	9*19*78 - 09*20*78	12/ 24	87	AEDC 16-FT TRANSONIC	517	2439	PUBLISHED
5/140C	.0300	OA146	11*28*78 - 12*07*78	80/116	30	ARC 8X7-FT SUPERSONIC	318	2445	PUBLISHED
5/140C	.0300	OA126C	12* 8*78 - 12*22*78	80/ 56	134	ARC 8X7-FT SUPERSONIC	289	2424	PUBLISHED
5/140C	.0300	IA184	4* 2*79 - 04*13*79	24/ 40	115	ARC 9X7-FT SUPERSONIC	347	2456	PUBLISHED
5/140C	.0300	IA190A	2* 7*80 - 02*21*80	160/168	166	ARC 11-FT TRANSONIC	411	2476	IN PROCESS
5/140C	.0300	OA400	4*23*80 - 05*02*80	120/120	200	ARC 11-FT TRANSONIC	427	2482	PUBLISHED
5/140C	.0300	IA190B	5*20*80 - 02*21*80	120/104	294	ARC 9X7-FT SUPERSONIC	411	2476	IN PROCESS

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 48

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
140C(MOD)/747	.0125	CA23A	3*21*75 - 04*17*75	120/213	71	ARC 14-FT TRANSONIC	085	2243	PUBLISHED
140C(MOD)/747	.0125	CA23B	5* 1*75 - 07*22*75	160/132	46	ARC 14-FT TRANSONIC	085	2275	PUBLISHED
140C(MOD)/747	.0125	CA26	8* 4*75 - 08*15*75	94/ 95	131	LTV 4X4-FT SUPERSONIC	559	2273	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 49

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
3A/140A	.0150	0A68	6•20•73 - 6•29•73	60/ 60	44	RI 7-FT TRISONIC	276	2061	PUBLISHED
3A/140A	.0150	0A23-1	7•19•73 - 07•31•73	80/ 54	23	ARC 3.5-FT HYPERSONIC	168	2071	PUBLISHED
4/140A.B	.0150	CA20A	9•10•73 - 09•13•73	50/ 40	29	LARC UNITARY PLAN	1057	2083	PUBLISHED
4/140A.B	.0150	0A25	9•14•73 - 09•21•73	80/ 88	156	LARC 8-FT TRANSONIC PRESSURE	661	2089	PUBLISHED
4/140A.B	.0150	0A87	10•15•73 - 10•23•73	80/ 80	30	ARC 3.5-FT HYPERSONIC	176	2115	PUBLISHED
4/140A/B	.0150	0A91	10•26•73 - 11•01•73	40/ 40	38	RI 7-FT TRISONIC	278	2116	PUBLISHED
4/140A.B	.0150	0A20C	11• 5•73 - 11•08•73	40/ 35	19	LARC UNITARY PLAN	1057	2147	PUBLISHED
4/140A.B	.0150	0A77	11•27•73 - 12•01•73	40/ 32	124	AEDC B / HYPERSONIC	474	2134	PUBLISHED
4/140A.B	.0150	0A78	12• 3•73 - 12•04•73	20/ 16	56	AEDC C / HYPERSONIC	474	2134	PUBLISHED
4/140A.B	.0150	0A84	12•10•73 - 12•14•73	80/115	207	LTV 4X4-FT SUPERSONIC	488	2037	PUBLISHED
4/140A.B	.0150	AA1A	2•12•74 - 02•25•74	64/ 64	28	ARC 3.5-FT HYPERSONIC	186		UNASSIGNED
4/140A.B	.0150	0A36	2•25•74 - 03•01•74	80/ 80	38	ARC 3.5-FT HYPERSONIC	187	2162	PUBLISHED
4/140A.B	.0150	0A59	3•13•74 - 3•21•74	120/293	150	ARC 6X6-FT SUPERSONIC	709	2159	PUBLISHED
4/140A/B	.0150	0A98	3•27•74 - 04•03•74	80/128	46	ARC 3.5-FT HYPERSONIC	190	2167	PUBLISHED
4/140A.B	.0150	AA1B	4• 4•74 - 04•06•74	64/ 64	13	ARC 3.5-FT HYPERSONIC	186		UNASSIGNED
4/140A.B	.0150	0A20B	4• 8•74 - 04•12•74	50/ 43	30	LARC UNITARY PLAN	1097	2163	PUBLISHED
4/140A.B	.0150	IA70	5• 3•74 - 05•24•74	80/161	173	RI 7-FT TRISONIC	282	2175	PUBLISHED
4/140A.B	.0150	0A116	6•10•74 - 06•14•74	80/ 80	81	LARC 8-FT TRANSONIC PRESSURE	686	2186	PUBLISHED
4/140A.B	.0150	IA110-1	7• 8•74 - 07•11•74	50/ 60	79	ARC 9X7-FT SUPERSONIC	052	2189	PUBLISHED
4/140A.B(MOD)	.0150	0A115A	7•29•74 - 07•31•74	24/ 28	82	AEDC A / SUPERSONIC	71A	2198	PUBLISHED
4/140A.B(MOD)	.0150	0A79	8• 1•74 - 08•03•74	24/ 23	79	AEDC B / HYPERSONIC	71A	2196	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 50

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL.	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
3A/139B	.0060	OH14	10•17•73 -	10•18•73	16/ 16	29	LARC MACH 8 VARIABLE DENSITY	648	2117	PUBLISHED
3A/139B	.0060	OH45	11• 2•73 -	11•09•73	40/ 46	22	LARC 20-IN FREON	121	2109	PUBLISHED
2A/089B	.0060	IH19A	12•14•73 -	12•26•73	40/ 40	22	LARC HYPERSONIC NITROGEN	28	2157	PUBLISHED
2A/089B	.0060	IH19B	12•27•73 -	01•08•74	20/ 40	22	LARC HYPERSONIC NITROGEN	28	2157	PUBLISHED
2A/089B	.0060	IH28-2	5•20•74 -	05•24•74	30/ 38	15	ARC 3.5-FT HYPERSONIC	195	2180	PUBLISHED
2A/089B	.0060	IH28-1	5•20•74 -	05•24•74	80/ 50	24	ARC 3.5-FT HYPERSONIC	195	2180	PUBLISHED
5/140C	.0060	IH73	12• 1•77 -	01•23•78	160/160	0	ARC 3.5-FT HYPERSONIC	233	2407	PUBLISHED

MODEL ID : 51

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140A,B	.0100	OA81	11*28*73 - 12*28*73	104/ 94	48	AEDC F / HYPERSONIC	489	2152	PUBLISHED
4/140A,B	.0100	OA113	8*10*74 - 10*04*74	24/336	108	CALSPAN HYPERSONIC SHOCK	184-	2234	PUBLISHED
4/140A,B	.0100	OA93	11*18*74 - 11*23*74	80/152	15	CALSPAN HYPERSONIC SHOCK	737	2238	PUBLISHED
4/140A,B	.0100	OA160	2* 5*75 - 02*08*75	12/ 12	14	AEDC F / HYPERSONIC	28A	2247	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 52

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139.089B	.0100	IA61B	2*26*74 - 02*26*74	8/ 8	9	AEDC A / SUPERSONIC	21AA	2226	PUBLISHED
3/139B	.0100	IA17A	3* 6*74 - 03*15*74	40/ 45	997	AEDC B / HYPERSONIC	422	2156	PUBLISHED
3/139B	.0100	IA17B	3*18*74 - 03*19*74	8/ 8	13	AEDC B / HYPERSONIC	422	2230	PUBLISHED
3/139B	.0100	IA18	4* 9*74 - 04*12*74	60/ 64	26	ARC 3.5-FT HYPERSONIC	191	2160	PUBLISHED
3/139B	.0100	IA87	7*18*74 - 07*20*74	24/ 23	90	AEDC A / SUPERSONIC	60A	2192	PUBLISHED
3/139B	.0100	IA111	3*21*75 - 03*28*75	36/ 33	***	AEDC A / SUPERSONIC	A3A	2242	PUBLISHED
5/140C	.0100	IA114	8*18*75 - 08*22*75	42/ 56	100	AEDC B / HYPERSONIC	C4A	2272	PUBLISHED

AUG 01,1984

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 53

REF. MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
FLAT PLATE	.1110	OH15	9•12•73	09•20•73	64/ 96	32	ARC 3.5-FT HYPERSONIC	173	2385	PUBLISHED
FLAT PLATE	.1110	OH44	10•24•73	10•30•73	80/ 80	46	ARC 3.5-FT HYPERSONIC	177	2386	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 54

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140B		.1400	056	9+ 2+74	- 09+12+74	120/104	27	LARC 16-FT TRANSONIC DYNAMIC	246	2365	PUBLISHED

AUG 01,1984

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 55

REF. MODEL	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140B	.1400	OS7	8*12*74 - 08*30*74	120/120	30	LARC 16-FT TRANSONIC DYNAMIC	246	2363	PUBLISHED
4/140A,B	.1400	OS22	4* 7*75 - 04*10*75	80/ 58	16	LARC 16-FT TRANSONIC DYNAMIC	258		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 56

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0175	OH25B	1*30*75 - 02*03*75	24/ 23	153	AEDC B / HYPERSONIC	83A	2366	PUBLISHED
5/140C	.0175	OH74	6* 3*75 - 06*12*75	12/ 12	0	AEDC B / HYPERSONIC	87A	2263	PUBLISHED
5/140C	.0175	IH42	4*26*76 - 05*26*76	192/218	57	ARC 3.5-FT HYPERSONIC	217		UNASSIGNED
5/140C	.0175	OH102A	10*25*78 - 11*29*78	8/ 13	0	AEDC B / HYPERSONIC	B65	2455	PUBLISHED
5/140C	.0175	IH102-2	5* 1*79 - 06*01*79	12/ 12	0	AEDC A / SUPERSONIC	B67	2464	PUBLISHED
5/140C	.0175	IH103-2	10*15*79 - 11*01*79	100/100	0	ARC 3.5-FT HYPERSONIC	245	2467	PUBLISHED
5/140C	.0175	OH109	10*27*80 - 11*24*80	48/ 40	0	AEDC B / HYPERSONIC	G9	2490	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 57

REF MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
VEH 101 (ADS)	.1000	0A220	11*11*75 -	11*21*75	120/110	142	ARC 14-FT TRANSONIC	150	2286	PUBLISHED
VEH 102 (ADS)	.1000	0A224	2*23*76 -	03*24*76	80/304	25	LARC 16-FT TRANSONIC	312	2329	PUBLISHED
VEH 102 (ADS)	.1000	0A228	5*29*76 -	05*01*76	16/ 23	45	RI 7X11-FT LOW SPEED	757	2322	PUBLISHED

MODEL ID : 58

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
FLAT PLATE	1.0000	IH51A	7*14*77 - 07*27*77	80/118	62	ARC 3.5-FT HYPERSONIC	228	2393	PUBLISHED
FLAT PLATE	1.0000	IH51B	7*15*78 - 07*24*78	60/ 60	0	ARC 3.5-FT HYPERSONIC	239	2429	PUBLISHED
FLAT PLATE	1.0000	IH51C	12*26*78 - 02*16*79	500/632	0	ARC 3.5-FT HYPERSONIC	241	2448	PUBLISHED
FLAT PLATE	1.0000	IH51D	5* 1*79 - 06*00*79	240/240	0	ARC 3.5-FT HYPERSONIC	244	2461	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 59

REF.	MODEL SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0100	IH43	12*17*75 - 02*23*76	120/250	60	CALSPAN HYPERSONIC SHOCK	18?	2319	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 60

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0175	OH39A	11*21*74 - 11*28*74	84/ 59	622	AEDC B / HYPERSONIC	74A	2241	PUBLISHED
5/140C	.0175	OH39B	1* 8*75 - 01*09*75	12/ 13	80	AEDC B / HYPERSONIC	74A	2241	PUBLISHED
5/140C	.0175	IH41A	3*31*75 - 5*21*75	48/ 57	318	AEDC A / SUPERSONIC	4A	2240	PUBLISHED
5/140C	.0175	IH48	4*17*75 - 05*08*75	240/256	125	ARC 3.5-FT HYPERSONIC	211	2248	PUBLISHED
5/140C	.0175	IH41B	12*11*75 - 01*09*76	78/ 80	300	AEDC A / SUPERSONIC	4A	2295	PUBLISHED
5/140C	.0175	IH47	3* 8*76 - 03*19*76	40/ 31	178	AEDC A / SUPERSONIC	J3A	2312	PUBLISHED
5/140C	.0175	OH98A	6*17*76 - 06*23*76	43/ 44	234	AEDC B / HYPERSONIC	J7A	2340	PUBLISHED
5/140C	.0175	OH98B	7*26*76 - 07*26*76	20/ 13	98	AEDC B / HYPERSONIC	J74	2340	PUBLISHED
5/140C	.0175	IH68	10*12*76 - 12*08*76	400/623	0	ARC 3.5-FT HYPERSONIC	222	2357	PUBLISHED
5/140C	.0175	IH72	1* 3*77 - 01*10*77	60/ 56	0	AEDC A / SUPERSONIC	K2A	2372	PUBLISHED
5/140C	.0175	OH84A-1	4*20*77 - 04*21*77	20/ 16	81	AEDC B / HYPERSONIC	R4A	2388	PUBLISHED
5/140C	.0175	IH90	1*30*78 - 03*10*78	160/116	73	ARC 3.5-FT HYPERSONIC	234	2412	PUBLISHED
5/140C	.0175	IH85	4*19*78 - 04*26*78	60/ 65	337	AEDC A / SUPERSONIC	W5	2431	PUBLISHED
5/140C	.0175	OH103B	4*27*78 - 04*28*78	24/ 12	53	AEDC D / HYPERSONIC	V2C	2427	PUBLISHED
5/140C	.0175	OH84B	5* 0*79 - 06*00*79	72/ 72	0	AEDC B / HYPERSONIC	B67	2464	PUBLISHED
5/140C	.0175	IH102-1	5* 1*79 - 06*01*79	26/ 26	0	AEDC A / SUPERSONIC	B67	2464	PUBLISHED
5/140C	.0175	OH105A	5*15*79 - 06*20*79	24/ 24	0	AEDC B / HYPERSONIC	B67	2464	PUBLISHED
5/140C	.0175	OH84C	6*15*79 - 06*28*79	80/ 80	0	ARC 3.5-FT HYPERSONIC	246	2468	PUBLISHED
5/140C	.0175	OH105B	7*23*79 - 08*01*79	24/180	0	ARC 3.5-FT HYPERSONIC	247	2468	PUBLISHED
5/140C	.0175	IH103-1	10* 1*79 - 11*01*79	100/100	0	ARC 3.5-FT HYPERSONIC	245	2467	PUBLISHED
5/140C	.0175	IH104	2* 7*80 - 04*17*80	80/ 80	0	ARC 3.5-FT HYPERSONIC	250	2480	PUBLISHED
5/140C	.0175	OH110	11*17*80 - 01*30*81	80/200	0	ARC 3.5-FT HYPERSONIC	253	2495	PUBLISHED
5/140C	.0175	OH111	9*24*81 - 09*30*81	32/ 32	0	AEDC B / HYPERSONIC	1C	2495	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 61

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140B		.0100	OH38	6*21*74 -	07*18*74	160/320	91	ARC 3.5-FT HYPERSONIC	198	2171	PUBLISHED

MODEL #62 WAS NOT BUILT..........

MODEL # 63 WAS NOT BUILT.............

MODEL ID : 64

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
3/139B	.0175	OH51-1	6*26*74 - 07*03*74	24/ 30	50	LARC 31-IN CONT-FLOW HYP.	112	2368	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

FEB 08, 1983

MODEL ID : 65

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0400	OH78	7- 2-76	11-24-76	480/ 1	0	JSC VAC. CHAMBER A	56-A	2371	PUBLISHED
5/140C	.0400	OH79	6- 1-78	08-24-78	288/288	0	JSC VAC. CHAMBER A	61-A	2443	PUBLISHED

FEB 08, 1983

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 66

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	STATUS
5/140C		.0250	OH66	8-30-76 - 10-17-76	120/120	30	CALSPAN HYPERSONIC SHOCK	131	2359	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 67

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140A.B	.0150	IA42A	11•27•73 - 12•04•73	40/ 70	62	LARC UNITARY PLAN	1056	2119	PUBLISHED
4/140A.B	.0150	IA41	12•11•73 - 12•14•73	80/ 64	86	LARC 8-FT TRANSONIC PRESSURE	667	2118	PUBLISHED
4/140A.B	.0150	IA42B	12•17•73 - 12•21•73	60/ 50	42	LARC UNITARY PLAN	1073	2119	PUBLISHED
4/140A.B	.0150	OA106	12•17•73 - 12•18•73	20/ 24	18	LARC 8-FT TRANSONIC PRESSURE	668	2120	PUBLISHED
4/140A.B	.0150	IA69	1•10•74 - 01•14•75	24/ 25	14	RI 7-FT TRISONIC	280	2122	PUBLISHED
4/140A.B	.0150	IA110-2	7• 8•74 - 07•11•74	30/ 20	17	ARC 9X7-FT SUPERSONIC	052	2189	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 68

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
ET FORETANK	.0700	IA137	4•26•76 - 05•03•76	40/ 56	43	ARC 14-FT TRANSONIC	143	2316	PUBLISHED
ET FORETANK	.0700	IA131B	11• 3•78 - 11•09•78	48/ 40	0	ARC 9X7-FT SUPERSONIC	283	2462	PUBLISHED
ET FORETANK	.0700	IA132	11•27•78 - 12•14•78	96/ 96	0	AEDC 16-FT TRANSONIC	505	2449	PUBLISHED
ET FORETANK	.0700	IA131C	3• 5•79 - 03•11•79	48/ 40	0	ARC 8X7-FT SUPERSONIC	283	2462	PUBLISHED
ET FORETANK	.0700	IA180	3•26•79 - 03•30•79	48/ 53	37	LARC UNITARY PLAN	1267	2457	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 69

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140A,B	.0150	LA71B	7*21*75 - 07*31*75	48/ 48	30	LARC UNITARY PLAN	1147	2271	PUBLISHED
4/140A,B	.0150	LA71A	10*17*75 - 10*22*75	48/ 64	15	LARC UNITARY PLAN	1132	2271	PUBLISHED
4/140A,B	.0150	LA73A	12*18*75 - 12*30*75	82/ 82	1	LARC LOW TURBULANCE PRESSURE	227	2298	PUBLISHED
4/140A,B	.0150	LA72	3*26*76 - 03*31*76	72/ 72	30	LARC 8-FT TRANSONIC PRESSURE	740	2309	PUBLISHED
4/140A,B	.0150	LA73B	12*10*76 - 12*13*75	16/ 16	6	LARC LOW TURBULANCE PRESSURE	238	2298	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 70

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0125	0A169	3*26*76 - 04*09*76	12/ 43	200	AEDC B / HYPERSONIC	D8A	2320	PUBLISHED
5/140C	.0125	IA22	5* 3*76 - 05*08*76	52/ 49	750	AEDC B / HYPERSONIC	59A	2327	PUBLISHED
5/140C	.0125	IA148	4*27*77 - 05*03*77	52/ 52	272	AEDC B / HYPERSONIC	TOA	2384	PUBLISHED
OV102	.0125	0A255A	10*13*80 - 11*07*80	240/228	268	LARC UNITARY PLAN	1311	2498	PUBLISHED
OV102	.0125	0A255B	11* 8*80 - 11*21*80	240/132	100	LARC UNITARY PLAN	1358	2498	PUBLISHED
OV102	.0125	0A255C	11*24*80 - 12*15*80	240/140	27	LARC UNITARY PLAN	1315	2498	PUBLISHED
OV102	.0125	0A255D	1*12*81 - 02*02*81	240/160	90	LARC UNITARY PLAN	1319	2498	PUBLISHED
OV102	.0125	0A256	2* 2*81 - 02*09*81	80/ 32	0	LARC 16-FT TRANSONIC	352		UNASSIGNED

MODEL # 71 WAS NOT BUILT..............

AUG 01, 1984

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 72

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	STATUS
4/140A,B	.0100	0A90	3* 4*74 - 03*06*74	25/ 40	43	LARC 31-IN CONT-FLOW HYP.	110	2149	PUBLISHED
4/140A,B	.0100	IA44A	8*12*74 - 08*16*74	40/ 50	27	LARC UNITARY PLAN	1088	2206	PUBLISHED
4/140A,B	.0100	IA44B	8*19*74 - 08*23*74	40/ 80	47	LARC UNITARY PLAN	1119	2206	PUBLISHED
4/140A,B	.0100	IA43	8*26*74 - 09*03*74	80/ 80	105	LARC 8-FT TRANSONIC PRESSURE	693	2204	PUBLISHED
4/140A,B	.0100	LA59	12*20*74 - 01*07*75	96/ 96	146	LARC 8-FT TRANSONIC PRESSURE	703	2233	PUBLISHED
4/140A,B	.0100	LA69	4*24*75 - 04*29*75	64/ 64	98	LARC 8-FT TRANSONIC PRESSURE	714	2257	PUBLISHED
5/140C	.0100	IA141	3*31*76 - 04*05*76	30/ 30	37	RI 7-FT TRISONIC	297	2315	PUBLISHED
5/140C	.0100	IA94A	4*18*76 - 04*23*76	40/ 60	92	LARC UNITARY PLAN	1152	2323	PUBLISHED
5/140C	.0100	IA94B	4*26*76 - 05*04*76	80/ 84	144	LARC UNITARY PLAN	1177	2324	PUBLISHED
5/140C	.0100	IA93	5*10*76 - 05*14*76	80/ 96	255	LARC 8-FT TRANSONIC PRESSURE	749	2326	PUBLISHED
5/140C	.0100	IA144	4* 6*77 - 04*15*77	160/200	514	ARC 11-FT TRANSONIC	228	2377	PUBLISHED
5/140C	.0100	IA244	5*24*77 - 06*01*77	80/ 76	154	LARC 8-FT TRANSONIC PRESSURE	779	2391	PUBLISHED
5/140C	.0100	LA113	8* 5*77 - 09*08*77	32/ 28	17	LARC 8-FT TRANSONIC PRESSURE	787	2397	PUBLISHED
VEH 102	.0100	LA141C	6*22*80 - 07*01*80	80/ 10	0	LARC 20-IN HYPERSONIC (M=6)	6550		UNASSIGNED
VEH 102	.0100	0A259	2*16*81 - 02*20*81	16/ 40	137	AEDC B / HYPERSONIC	14	2493	PUBLISHED
VEH 102	.0100	0A257	3*12*81 - 04*20*81	80/324	380	LARC 20-IN HYPERSONIC (M=6)	6559	2466	PUBLISHED
VEH 102	.0100	IA193	2*26*82 - 04*31*82	72/720	0	AEDC A / SUPERSONIC	A1G		UNASSIGNED

MODEL # 73 WAS NOT BUILT.............

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 74

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0040	IA33	5* 9*74 - 07*21*74	256/264	270	MSFC 14-IN TRANSONIC	594	2174	PUBLISHED
5/140C	.0040	OA108	6*24*74 - 07*09*74	80/ 80	186	MSFC 14-IN TRANSONIC	599	2190	PUBLISHED
5/140C	.0040	OA89	7*15*74 - 08*05*74	60/143	32	LARC HYPERSONIC NITROGEN	30	2214	PUBLISHED
5/140C	.0040	OA109	8*26*74 - 08*29*74	60/ 88	32	LARC 22-IN HELIUM	431	2205	PUBLISHED
5/140C	.0040	OA131	9*11*74 - 09*26*74	80/ 96	109	MSFC 14-IN TRANSONIC	607	2232	PUBLISHED
5/140C	.0040	IA71A-2	12*11*74 - 12*17*74	20/ 17	29	MSFC 14-IN TRANSONIC	610	2227	PUBLISHED
5/140C	.0040	IA71B-2	12*19*74 - 01*09*75	16/ 16	41	MSFC 14-IN TRANSONIC	610	2227	PUBLISHED
5/140C	.0040	FA14	1* 9*75 - 07*06*75	60/142	0	MSFC 14-IN TRANSONIC	600	2274	PUBLISHED
5/140C	.0040	IA125-1	4*25*75 - 05*22*75	60/ 93	137	MSFC 14-IN TRANSONIC	622	2253	PUBLISHED
5/140C	.0040	IA140A	6* 1*76 - 08*03*76	64/222	230	MSFC 14-IN TRANSONIC	641	2335	PUBLISHED
5/140C	.0040	IA140B	10* 1*76 - 01*28*77	80/279	44	MSFC 14-IN TRANSONIC	646	2335	PUBLISHED
5/140C	.0040	LA124	6* 7*77 - 06*10*77	40/ 40	19	LARC UNITARY PLAN	1207	2426	PUBLISHED
5/140C	.0040	IA181	12*15*77 - 02*03*78	120/120	111	MSFC 14-IN TRANSONIC	649	2406	PUBLISHED
5/140C	.0040	FA25	4*15*78 - 08*01*78	200/294	0	MSFC 14-IN TRANSONIC	652	2437	PUBLISHED
5/140C	.0040	FA26	5* 1*78 - 06*01*78	80/ 80	0	MSFC 14-IN TRANSONIC	653		UNASSIGNED
5/140C	.0040	FA27	3*14*79 - 05*16*79	150/160	0	MSFC 14-IN TRANSONIC	655	2460	IN PROCESS
5/140C	.0040	FA29	4* 1*79 - 05*00*79	400/ 0	0	MSFC 14-IN TRANSONIC	657		UNASSIGNED
5/140C	.0040	FA28	8* 1*79 - 09*01*79	200/ 0	0	MSFC 14-IN TRANSONIC	656	2474	PUBLISHED
VEH. 102	.0040	LA141A	1*12*80 - 02*01*80	80/148	0	LARC 20-IN HYPERSONIC (M=6)	6546	2477	PUBLISHED
VEH. 102	.0040	LA142	2* 1*80 - 03*01*80	80/ 80	0	LARC 20-IN FREON	390		UNASSIGNED
VEH 102	.0040	LA141B	3*18*80 - 05*01*80	80/200	0	LARC 20-IN HYPERSONIC (M=6)	6546	2477	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 75

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0100	IA82C	11*11*74	11*15*74	80/ 92	240	ARC 8X7-FT SUPERSONIC	044	2219	PUBLISHED
5/140C	.0100	IA82B	1*28*75	02*04*75	70/132	286	ARC 9X7-FT SUPERSONIC	044	2231	PUBLISHED
5/140C	.0100	IA40	6*23*76	06*29*76	26/ 41	346	AEDC A / SUPERSONIC	425	2293	PUBLISHED
5/140C	.0100	IA142	8*11*76	08*18*76	78/ 64	***	AEDC A / SUPERSONIC	K1A	2346	PUBLISHED
5/140C	.0100	IA143	11* 8*76	11*13*76	65/ 58	***	AEDC A / SUPERSONIC	P8A	2354	PUBLISHED
5/140C	.0100	IA138	8*21*78	09*01*78	70/112	224	ARC 9X7-FT SUPERSONIC	246	2438	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 76

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
OV101(ALT)	.3600	OA100	5*27*75	06*14*75	240/272	190	ARC 40X80-FT SUBSONIC	462	2261	PUBLISHED
OV101(ALT)	.3600	OA164	11*28*75	12*01*75	80/ 80	22	ARC 40X80-FT SUBSONIC	473	2499	PUBLISHED
OV101(ALT)	.3600	OA174	2* 2*76	02*27*76	240/264	165	ARC 40X80-FT SUBSONIC	479	2302	PUBLISHED

MODEL ID : 77

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0040	IA71A-1	12*11*74 -	12*17*74	20/ 40	40	MSFC 14-IN TRANSONIC	610	2227	PUBLISHED
5/140C	.0040	IA71B-1	12*19*74 -	01*09*75	40/ 64	90	MSFC 14-IN TRANSONIC	610	2227	PUBLISHED
5/140C(74TS)	.0040	IA125-2	4*25*75 -	05*22*75	40/ 30	50	MSFC 14-IN TRANSONIC	622	2253	PUBLISHED

MODEL # 78 WAS NOT BUILT...........

MODEL ID : 79

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C		.0550	0S20	10*22*75 -	10*30*75	120/120	14	LARC 16-FT TRANSONIC DYNAMIC	266		UNASSIGNED

AUG 01, 1984

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 80

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL	NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	FLUTTER	.0550	OS21	5* 8*78 -	05*26*78	200/120	0	LARC 16-FT	TRANSONIC DYNAMIC	300		UNASSIGNED
5/140C	FLUTTER	.0550	IS10	9*18*78 -	10*08*78	200/128	0	LARC 16-FT	TRANSONIC DYNAMIC	308		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 81

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
HRSI TILE	1.0000	OS8A	7*11*74 -	07*18*74	60/156	96	ARC 11-FT TRANSONIC	705	2179	PUBLISHED
HRSI TILE	1.0000	OS8B	7*19*74 -	07*29*74	60/120	39	ARC 9X7-FT SUPERSONIC	705	2179	PUBLISHED
HRSI TILE	1.0000	OS37	5* 7*79 -	05*11*79	60/ 40	0	ARC 9X7-FT SUPERSONIC	369	2458	PUBLISHED
TILE	1.0000	OS55	2*23*81 -	03*02*81	80/ 64	0	ARC 9X7-FT SUPERSONIC	464	2465	PUBLISHED
TILE	1.0000	OS57	8*26*81 -	08*27*81	8/ 8	0	ARC 9X7-FT SUPERSONIC	508	2465	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 82

REF.	MODEL SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0400	OH50A	3*29*74 - 04*11*74	8/ 16	66	AEDC B / HYPERSONIC	526	2285	PUBLISHED
5/140C	.0400	OH54A	10* 4*74 - 10* 8*74	36/ 32	117	AEDC B / HYPERSONIC	82A	2301	PUBLISHED
5/140C	.0400	OH54B	7*21*75 - 07*25*75	48/ 52	124	AEDC B / HYPERSONIC	82A	2342	PUBLISHED
5/140C	.0400	OH54C	8*26*75 - 09*02*75	48/ 48	120	AEDC B / HYPERSONIC	82A	2342	PUBLISHED
5/140C	.0400	OH75	9* 2*75 - 09*03*75	14/ 13	44	AEDC B / HYPERSONIC	E3A	2303	PUBLISHED
5/140C	.0400	OH69	11*14*75 - 12*11*75	84/ 87	246	AEDC B / HYPERSONIC	E9A	2321	PUBLISHED
5/140C	.0400	OH53B	4*14*76 - 04*23*76	40/ 80	23	ARC 3.5-FT HYPERSONIC	216	2317	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 83

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0400	0H50B	7*12*74 - 07*17*74	36/ 27	220	AEDC B / HYPERSONIC	58A	2358	PUBLISHED
5/140C	.0400	0H60	5*12*75 - 05*12*75	12/ 12	139	AEDC B / HYPERSONIC	B7A	2356	PUBLISHED
5/140C	.0400	0H53A	4* 7*76 - 04*13*76	40/ 40	39	ARC 3.5-FT HYPERSONIC	216	2317	PUBLISHED
5/140C	.0400	0H84A-2	4*20*77 - 04*21*77	5/ 9	16	AEDC B / HYPERSONIC	R4A	2388	PUBLISHED
VEH. 5 F'BODY	.0400	0H103A	2*20*78 - 02*21*78	12/ 8	72	AEDC B / HYPERSONIC	V2C	2420	PUBLISHED
5/140C	.0400	IH102-3	5* 1*79 - 06*01*79	10/ 10	0	AEDC A / SUPERSONIC	B67	2464	PUBLISHED

AUG 01,1984

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 84

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0350	IS2B	9*25*75 - 10*29*75	60/ 60	67	ARC 9X7-FT SUPERSONIC	113	2284	PUBLISHED
5/140C	.0350	IS2A	11* 7*75 - 11*14*75	144/120	53	ARC 11-FT TRANSONIC	113	2284	PUBLISHED
5/140C	.0350	IH11	4* 1*78 - 04*18*78	80/ 64	0	LERC 10X10-FT SUPERSONIC	045	2428	PUBLISHED
5/140C	.0350	OA253	7* 1*80 - 07*08*80	80/ 80	139	AEDC 16-FT TRANSONIC	574	2486	PUBLISHED

MODEL ID : 85

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
LRSI TILE	1.0000	OS13	11*24*75 - 11*26*75	16/ 21	45	ARC 9X7-FT SUPERSONIC	166	2287	IN PROCESS
LRSI TILE	1.0000	OS12	1*11*76 - 01*29*76	80/ 40	42	ARC 2X2-FT TRANSONIC	116	2450	PUBLISHED

MODEL # 86 WAS NOT BUILT..........

MODEL # 87 WAS NOT BUILT.........

AUG 01.1984

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 88

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C	.0200	IA19A	9*16*74	09*23*74	156/136	201	ARC 11-FT TRANSONIC	014	2170	PUBLISHED
5/140C	.0200	IA80	11* 4*74	11*08*74	100/144	380	ARC 11-FT TRANSONIC	023	2212	PUBLISHED
5/140C	.0200	IA72	5*19*75	05*31*75	120/200	176	ARC 11-FT TRANSONIC	072	2258	PUBLISHED
140C/747	.0200	LA80	10* 6*75	11*07*75	156/156	83	LARC 7X10-FT HIGH SPEED	999	2299	PUBLISHED
5/140C	.0200	IA119	10* 7*77	10*31*77	170/285	620	ARC 11-FT TRANSONIC	275	2404	PUBLISHED

MODEL ID : 89

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL	NO.	DOCUMENT NO.	DOCUMENT STATUS
VEH 102	.0200	IA156A	10*28*77	11*10*77	96/124	575	AEDC 16-FT	TRANSONIC	470	2403	PUBLISHED
VEH 102	.0200	IA156B	12*16*77	01*06*78	100/191	177	ARC 9X7-FT	SUPERSONIC	272	2408	PUBLISHED
VEH 102	.0200	IA183	11*15*78	11*16*78	12/ 12	40	AEDC 16-FT	TRANSONIC	519	2444	PUBLISHED
VEH 102	.0200	LA132	10*11*79	11*01*79	40/ 80	18	LARC 16-FT	TRANSONIC	341	2471	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 90

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
4/140B		.0060	OH46	11*12*73 - 12*07*73	40/ 72	100	LARC MACH 8 VARIABLE DENSITY	4556	2350	PUBLISHED
4/140B		.0060	OH51-3	6*26*74 - 07*03*74	12/100	100	LARC 31-IN CONT-FLOW HYP.	112	2368	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 91

REF.	MODEL SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
WING TIP SEAL	.0800	OH56	12* 6*77 -	12*10*77	48/ 36	255	AEDC B / HYPERSONIC	R3A	2410	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 92

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	STATUS
140C	.0175	OH57A	10* 6*76 - 20*06*76	13/ 11	40	AEDC B / HYPERSONIC	K3A	2367	PUBLISHED
140C	.0175	OH57B	12* 4*76 - 12*05*76	26/ 34	14	AEDC B / HYPERSONIC	K3A	2367	PUBLISHED
140C SILTS	.0175	OH400	8* 1*79 - 09*01*79	36/ 36	124	ARC 11-FT TRANSONIC	B65	2472	PUBLISHED

MODEL ID : 93

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
ELEV/ELEV SEAL	.0100	OH58	3*24*78 - 04*21*78	120/168	58	ARC 3.5-FT HYPERSONIC	235	2417	PUBLISHED
ELEV/ELEV	.0100	OH108	12*15*80 - 01*15*81	200/139	43	ARC 3.5-FT HYPERSONIC	254	2494	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 94

MODEL REF.	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
ELEV/ELEV	.0250	OH90	3* 2*78	03*11*78	48/ 64	162	AEDC B / HYPERSONIC	P4A	2451	PUBLISHED
ELEV/ELEV	.0250	OH107	1* 7*81	01*08*81	12/ 12	0	AEDC B / HYPERSONIC	B17	2492	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 95

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
JSC	040A ORB.	.0500	MA1	8*25*72	09*06*72	80/ 80	120	LTV 15X20-FT LOW SPEED	407	2004	PUBLISHED
JSC	040A ORB.	.0500	MA8	12*15*72	01*27*73	40/ 40	40	TEXAS A+M 7X10-FT LOW SPEED	MA8		UNASSIGNED
2A/089B(CAN)		.0500	MA14	4*23*73	05*02*73	80/ 62	103	LTV 15X20-FT LOW SPEED	422	2283	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 96

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
LRSI TILES	1.0000	OS31	11*22*77 - 11*30*77	84/ 56	55	ARC 11-FT TRANSONIC	145		PUBLISHED
HRSI TILE	1.0000	OS36	4*16*79 - 04*19*79	60/ 64	0	ARC 11-FT TRANSONIC	369	2458	PUBLISHED
LRSI TILES	1.0000	OS41	4*18*79 - 04*20*79	8/ 16	0	ARC 11-FT TRANSONIC	369	2458	PUBLISHED
TPS TILES	1.0000	OS42	7* 2*79 - 07*05*79	8/ 8	0	ARC 11-FT TRANSONIC	380	2463	PUBLISHED
TPS TILES	1.0000	OS45	9* 3*79 - 09*03*79	8/ 8	0	ARC 11-FT TRANSONIC	381	2470	PUBLISHED
TPS TILE	1.0000	OS51B	11*17*80 - 11*19*80	50/ 50	0	ARC 11-FT TRANSONIC	436	2487	PUBLISHED
TPS TILE	1.0000	OS51A	11*17*80 - 01*29*81	50/ 77	0	ARC 11-FT TRANSONIC	436	2487	PUBLISHED
TPS TILE	1.0000	OS51C	1*27*81 - 01*29*81	27/ 27	0	ARC 11-FT TRANSONIC	436	2487	PUBLISHED
TPS TILE	1.0000	OS60	6* 9*81 - 06*09*81	8/ 8	0	ARC 11-FT TRANSONIC	500	2506	PUBLISHED

MODEL #97 WAS NOT BUILT

PHASE C/D WIND TUNNEL TESTING PER MUDEL

AUG 01,1984

MODEL ID : 98

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C		.0100	IH99	8*28*77	09*07*77	80/ 79	0	ARC 3.5-FT HYPERSONIC	230	2452	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01, 1984

MODEL ID : 99

REF. MODEL	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
ADS PROBES	1000	OA236	5*28*76 - 06*02*76	10/ 37	204	RI 7X11-FT LOW SPEED	759	2337	PUBLISHED
ADS PROBES	1000	OA238	10*25*76 - 11*08*76	24/ 48	57	RI 7X11-FT LOW SPEED	764	2351	PUBLISHED
ADS PROBES	1000	OA221B	11* 8*76 - 21*15*76	60/ 76	184	ARC 9X7-FT SUPERSONIC	119	2360	PUBLISHED
ADS PROBES	1000	OA221C	11*15*76 - 11*22*76	60/ 68	58	ARC 8X7-FT SUPERSONIC	119	2360	PUBLISHED
ADS PROBES	.1000	OA237	1*24*77 - 01*31*77	60/ 60	32	ARC 40X80-FT SUBSONIC	500	2375	PUBLISHED
ADS PROBES	.1000	OA234	6* 7*77 - 08*11*77	80/ 80	63	LERC 10X10-FT SUPERSONIC	042	2400	PUBLISHED
ADS PROBES	.1000	OA232	2*17*78 - 03*01*78	80/ 80	281	AEDC 16-FT TRANSONIC	431	2414	PUBLISHED
ADS PROBES	.1000	OA251B	4*17*78 - 04*23*78	40/ 80	90	ARC 9X7-FT SUPERSONIC	282	2421	PUBLISHED
ADS PROBES	.1000	OA251C	5*29*78 - 06*15*78	40/ 72	96	ARC 8X7-FT SUPERSONIC	282	2421	PUBLISHED
ADS PROBES	.1000	MA34	3*12*81 - 03*20*81	40/ 60	0	AEDC 16-FT TRANSONIC	594	2497	IN PROCESS
ADS PROBES	.1000	MA35B	12* 2*81 - 12*16*81	40/ 80	0	ARC 9X7-FT SUPERSONIC	513		UNASSIGNED
ADS PROBES	.1000	MA35C	4*19*82 - 04*23*82	40/120	0	ARC 8X7-FT SUPERSONIC	513		UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 100

REF. MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
5/140C + TOWER	.0460	IS20	7*24*78 -	08*25*78	200/216	0	LARC 16-FT TRANSONIC DYNAMIC	306		UNASSIGNED

MODEL # 101 WAS NOT BUILT.........

MODEL # 102 WAS NOT BUILT.....

C-5

MODEL # 103 WAS NOT BUILT.........

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 104

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
VEH 102	.0200	OA270C	4* 8*78 - 04*28*78	20/ 72	80	LARC 16-FT TRANSONIC	325	2419	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

AUG 01,1984

MODEL ID : 105

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	STATUS
VEH. 102	.0200	OA209	3*21*78 - 03*30*78	65/ 69	324	AEDC A / SUPERSONIC	P5A	2415	PUBLISHED
VEH. 102	.0200	OA208	3*30*78 - 04*06*78	52/ 47	183	AEDC B / HYPERSONIC	P5A	2416	PUBLISHED
VEH. 102	.0200	OA270B	5* 1*78 - 05*12*78	40/ 80	357	LARC 16-FT TRANSONIC	325	2419	PUBLISHED
VEH. 102	.0200	OA171	6* 5*78 - 06*22*78	180/180	35	NSWC HYPERSONIC LAB (#9)	1310	2433	PUBLISHED
VEH. 102	.0200	LA125	7* 3*78 - 07*05*78	16/ 48	41	LARC UNITARY PLAN	1243	2432	PUBLISHED
VEH. 102	.0200	LA140	12*26*79 - 01*03*80	80/ 80	17	LARC 16-FT TRANSONIC	342	2475	PUBLISHED

MODEL ID : 106

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
VEH 102	.0200	LA143	12*21*79 - 01*08*80	80/ 88	0	LARC 8-FT TRANSONIC PRESSURE	865		UNASSIGNED
VEH 102	.0200	LA131	1* 8*80 - 02*01*80	80/144	624	LARC UNITARY PLAN	1299	2478	PUBLISHED
VEH 102	.0200	LA144	7*28*80 - 08*01*80	80/138	198	LTV 4X4-FT SUPERSONIC	742	2484	PUBLISHED
VEH 102	.0200	OA258	11*25*80 - 01*06*81	48/128	541	AEDC B / HYPERSONIC	BHO	2491	PUBLISHED
VEH 102	.0200	MA37	11* 2*81 - 11*04*81	40/ 24	100	LARC UNITARY PLAN	1394		UNASSIGNED
VEH 102	.0200	MA33A	4*19*82 - 04*30*82	80/144	0	ARC 11-FT TRANSONIC	510	2507	PUBLISHED
VEH 102	.0200	MA33B	5*10*82 - 05*21*82	40/ 96	0	ARC 9X7-FT SUPERSONIC	510	2507	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 107

REF.	MODEL SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
TILE	.3300	0A252	10*16*79 - 11*14*79	360/360	0	ARC 2X2-FT TRANSONIC	382	2473	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

FEB 08,1983

MODEL ID : 108

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
TPS	1.0000	OS46B	3* 6*80 - 03*08*80	24/ 24	0	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
TPS	1.0000	OS46D	8* 0*80 - 09*00*80	8/ 8	0	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
TPS	1.0000	OS46E	9*15*80 - 10*20*80	8/ 8	0	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
TPS	1.0000	OS46F	10* 1*80 - 10*02*80	8/ 8	0	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
TPS TILE	1.0000	OS56	8*26*81 - 08*27*81	8/ 8	0	AEDC 16-FT TRANSONIC	608	2489	PUBLISHED
TPS	1.0000	OS46G	12*10*81 - 12*11*81	8/ 17	0	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED

MODEL ID : 103

REF.	MODEL SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
TPS	1.0000	OS46A	3* 4*80 - 03*05*80	8/ 8	0	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED
TPS	1.0000	OS46C	4*17*80 - 04*18*80	8/ 8	0	AEDC 16-FT TRANSONIC	551	2505	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

FEB 08,1983

MODEL ID : 110

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED.	TESTING COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
TPS		1.0000	OS47	11* 1*79 -	12*20*79	160/160	0	AEDC 1-FT TRANSONIC			UNASSIGNED

PHASE C/D WIND TUNNEL TESTING PER MODEL

MODEL ID : 111

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
TPS		1.0000	0549	1*28*81 - 02*04*81	40/ 44	0	AEDC 16-FT TRANSONIC	556	2483	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

FEB 08,1983

MODEL ID : :12

MODEL REF.	SCALE	TEST NO.	TESTING SCHED. COMPL	HOURS EST/CHG	RUNS	FACILITY	WIND TUNNEL NO.	DOCUMENT NO.	DOCUMENT STATUS
FUEL LINE	.5000	IA191	6+20+80 - 06+27+80	40/ 40	0	ARC 11-FT TRANSONIC	412	2378	PUBLISHED

PHASE C/D WIND TUNNEL TESTING PER MODEL

FEB 08,1983

MODEL ID : 113

REF.	MODEL	SCALE	TEST NO.	TESTING SCHED. - COMPL	HOURS EST/CHG	RUNS	WIND TUNNEL FACILITY	NO.	DOCUMENT NO.	DOCUMENT STATUS
TPS		1.0000	0S50	4*30*81 - 06*01*81	40/113	0	ARC 11-FT TRANSONIC	425	2485	PUBLISHED

APPENDIX B
DOCUMENTATION LISTING

APPENDIX B

TABLE B1. - DATAMAN LISTINGS WITH TEST, MODEL, FACILITY DATA

PHASE C/D SSV WIND TUNNEL TESTING DATAMAN LISTING

DATAMAN		TEST	TESTING		MODEL		WIND TUNNEL	
NO.	STATUS	NO.	SCHED.	COMPL.	REF.	(ID)	FACILITY	NO.
2001	PUBLISHED	MA5	9*15*72 -	09*25*72	PRE-ATP/001	(10)	LARC UNITARY PLAN	1002
2002	PUBLISHED	LA1	11*19*72 -	12*19*72	ATP	(6)	LARC 8-FT TRANSONIC PRESSURE	626
2003	PUBLISHED	MA2	9*18*72 -	11*06*72	ATP	(1)	LARC 22-IN HELIUM	409
2004	PUBLISHED	MA1	8*25*72 -	09*06*72	JSC 040A ORB.	(95)	LTV 15X20-FT LOW SPEED	407
2005	PUBLISHED	OA1	9*27*72 -	10*07*72	ATP	(1)	MSFC 14-IN TRANSONIC	555
2006	PUBLISHED	IA1A	10*10*72 -	10*19*72	ATP	(1)	MSFC 14-IN TRANSONIC	556
2007	PUBLISHED	OA4	10* 2*72 -	10*17*72	ATP	(6)	ARC 3.5-FT HYPERSONIC	147
2008	PUBLISHED	MA4	10* 1*72 -	10*02*72	RI ATP ORBITER	(0)	LARC 31-IN CONT-FLOW HYP.	089
2009	PUBLISHED	OA3	10*24*72 -	11*10*72	ATP	(6)	ARC 6X6-FT SUPERSONIC	650
2010	PUBLISHED	IA1B	10*19*72 -	11*28*72	ATP	(1)	MSFC 14-IN TRANSONIC	545
2011	PUBLISHED	MA9F	11*29*72 -	12*07*72	ATP	(1)	MSFC 14-IN TRANSONIC	558
2012	PUBLISHED	SA1F	12* 9*72 -	12*23*72	PRR/SRB	(1)	MSFC 14-IN TRANSONIC	554
2013	PUBLISHED	IA2	10*11*72 -	11*03*72	PRE-ATP/001	(7)	ARC 9X7-FT SUPERSONIC	616
2014	PUBLISHED	OA7	11*27*72 -	12*08*72	ATP	(6)	LARC UNITARY PLAN	1007
2015	PUBLISHED	IA4	11* 2*72 -	11*17*72	PRE-ATP/001	(9)	LTV 4X4-FT SUPERSONIC	458
2016	PUBLISHED	OA2	9*25*72 -	10*03*72	ATP	(2)	RI 7X11-FT LOW SPEED	689
2017	PUBLISHED	OA5	10*11*72 -	10*19*72	ATP-MODIF	(2)	RI 7X11-FT LOW SPEED	690
2018	PUBLISHED	IA3	11* 3*72 -	11*16*72	PRE-ATP/001	(10)	RI 7X11-FT LOW SPEED	693
2019	PUBLISHED	OA6	11*16*72 -	12*06*72	PRR	(2)	RI 7X11-FT LOW SPEED	694
2020	PUBLISHED	OA9	12*18*72 -	01*09*73	2A/089B	(2)	RI 7X11-FT LOW SPEED	696

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2021	PUBLISHED	OA45	2*21*73 -	02*28*73	2A/089B	(2)	RI 7X11-FT LOW SPEED	699
2022	PUBLISHED	OA10	1*30*73 -	02*16*73	2A/089B	(2)	RI 7X11-FT LOW SPEED	698
2023	PUBLISHED	LA2	10* 6*72 -	12*07*72	L/O-100 ORB.	(0)	LARC 22-IN HELIUM	411
2024	PUBLISHED	IA7	2*12*73 -	02*23*73	PRE-ATP/001	(7)	ARC 11-FT TRANSONIC	686
2025	PUBLISHED	SA3F	2*20*73 -	03*20*73	SRB	(449)	MSFC 14-IN TRANSONIC	565
2026	PUBLISHED	IA31FA	4* 9*73 -	04*13*73	2A/089B	(13)	MSFC 14-IN TRANSONIC	566
2027	PUBLISHED	IA32F	5* 9*73 -	05*24*73	2A/089B	(13)	MSFC 14-IN TRANSONIC	567
2028	PUBLISHED	IA31FB	4*13*73 -	04*30*73	2A/089B	(13)	MSFC 14-IN TRANSONIC	570
2029	PUBLISHED	OA47	3*28*73 -	04*05*73	2A/089B	(13)	MSFC 14-IN TRANSONIC	568
2030	PUBLISHED	OA14	2*28*73 -	03*15*73	2A/089B	(2)	RI 7X11-FT LOW SPEED	700
2031	PUBLISHED	LA3	8*23*72 -	11*16*72	L/O-100 ORB.	(0)	LARC 31-IN CONT-FLOW HYP.	085
2032	PUBLISHED	IA9C	4*22*73 -	05*01*73	2A/089B	(17)	ARC 8X7-FT SUPERSONIC	707
2033	PUBLISHED	LA4C	2*19*73 -	02*23*73	L/O-100 ORB.	(0)	LARC UNITARY PLAN	995
2034	PUBLISHED	LA22	6*19*72 -	06*30*72	JSC 049	(0)	LARC 22-IN HELIUM	405
2035	PUBLISHED	OH2	4*18*73 -	06*01*73	TPS TILES	(15)	ARC 3.5-FT HYPERSONIC	158
2036								
2037	PUBLISHED	OA84	12*10*73 -	12*14*73	4/140A,B	(49)	LTV 4X4-FT SUPERSONIC	488
2038	PUBLISHED	OA16	3*19*73 -	04*17*73	2A/089B	(2)	RI 7X11-FT LOW SPEED	701
2039	PUBLISHED	IA6	4*30*73 -	05*03*73	2A/089B	(13)	MSFC 14-IN TRANSONIC	571
2040	PUBLISHED	LA6	4*12*73 -	04*18*73	089B, 139 NOSE	(0)	LARC 8-FT TRANSONIC PRESSURE	643
2041								
2042	PUBLISHED	IA52	10*11*73 -	10*17*73	3A/139B	(34)	MSFC 14-IN TRANSONIC	584
2043	PUBLISHED	LA16	6*26*72 -	08*23*72	HRSI TILE	(0)	LARC MACH 8 VARIABLE DENSITY	624
2044	PUBLISHED	OA11A	4* 9*73 -	04*17*73	2A/089B	(18)	ARC 3.5-FT HYPERSONIC	157
2045	PUBLISHED	OA18	5* 8*73 -	05*17*73	3/139B	(43)	RI 7X11-FT LOW SPEED	704

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2046	PUBLISHED	LA17	5* 4*73	05*14*73	L/O-100 ORB	(0)	LARC 8-FT TRANSONIC PRESSURE	648
2047	PUBLISHED	LA31	8* 9*73	08*16*73	LARC ORB	(0)	LARC 31-IN CONT-FLOW HYP.	098
2048	PUBLISHED	IA12B	4*23*73	05*07*73	2A/089B(MOD)	(14)	ARC 9X7-FT SUPERSONIC	710
2049	PUBLISHED	OH40	1*30*73	02*05*73	2A/089B	(31)	LARC MACH 8 VARIABLE DENSITY	3619
2050	PUBLISHED	OA43	4*18*73	05*04*73	2A/089B	(18)	ARC 6X6-FT SUPERSONIC	706
2051	PUBLISHED	SA5F	5* 3*73	05*08*73	SRB	(449)	MSFC 14-IN TRANSONIC	572
2052								
2053	PUBLISHED	OA21B	6*21*73	06*25*73	3/139B W/CANS	(43)	RI 7X11-FT LOW SPEED	705
2054	PUBLISHED	LA8C	7* 3*73	07*06*73	089B, 139 NOSE	(0)	LARC UNITARY PLAN	1039
2055	PUBLISHED	OA48	5*25*73	6*11*73	3/139B,W/CANS	(34)	MSFC 14-IN TRANSONIC	574
2056	PUBLISHED	LA9C	10*31*73	11*02*73	089B,.139 NOSE	(0)	LARC LOW TURBULANCE PRESSURE	148
2057	PUBLISHED	OA44-2	6*11*73	06*15*73	3/139B	(42)	LARC UNITARY PLAN	1035
2058	PUBLISHED	OA17-2	6*18*73	07*06*73	2A/089B	(18)	LARC LOW TURBULANCE PRESSURE	138
2059	PUBLISHED	OA11B	5*14*73	05*25*73	2A/089B	(18)	ARC 3.5-FT HYPERSONIC	160
2060	PUBLISHED	OA58	6* 4*73	06*18*73	3/139B	(42)	ARC 3.5-FT HYPERSONIC	163
2061	PUBLISHED	OA68	6*20*73	6*29*73	3A/140A	(49)	RI 7-FT TRISONIC	276
2062	PUBLISHED	IA13	7* 5*73	07*17*73	3/139B	(32)	AEDC A / SUPERSONIC	323
2063	PUBLISHED	IA48	7*18*73	07*21*73	3A/139B	(34)	MSFC 14-IN TRANSONIC	580
2064	PUBLISHED	IA36	6*15*73	06*22*73	2A/089(MOD)	(14)	CALSPAN 8-FT TRANSONIC	053
2065	PUBLISHED	IA12C	7*11*73	07*27*73	2A/089(MOD)	(14)	ARC 8X7-FT SUPERSONIC	710
2066	PUBLISHED	LA11	7*11*73	07*20*73	089B, 139 NOSE	(0)	LARC 31-IN CONT-FLOW HYP.	096
2067	PUBLISHED	OS2	6* 4*73	06*07*73	ATP	(24)	LARC 26-IN TRANSONIC BLOWDOWN	544
2068	PUBLISHED	OA71A	7*27*73	08*03*73	2A/089B	(2)	RI 7X11-FT LOW SPEED	708
2069	PUBLISHED	MA7	5*14*73	05*18*73	2A/089B	(6)	LARC UNITARY PLAN	1031
2070	PUBLISHED	LA23	7*31*73	08*03*73	L/O-100 ORB.	(0)	LARC LOW TURBULANCE PRESSURE	141

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2071	PUBLISHED	OA23-2	7*26*73 -	07*31*73	3/139B	(32)	ARC 3.5-FT HYPERSONIC	168
2072	PUBLISHED	IA31FC	6*21*73 -	07*09*73	2A/089B	(13)	MSFC 14-IN TRANSONIC	573
2073	PUBLISHED	OA70	7*20*73 -	7*26*73	3/139B	(42)	LARC UNITARY PLAN	1043
2074	PUBLISHED	OA57A	8* 6*73 -	8*17*73	2A/089B	(2)	RI 7X11-FT LOW SPEED	709
2075	PUBLISHED	OH41A	3*19*73 -	03*28*73	2A/089B	(33)	LARC MACH 8 VARIABLE DENSITY	3778
2076	PUBLISHED	OH41B	5* 8*73 -	05*10*73	2A/089B	(38)	LARC MACH 8 VARIABLE DENSITY	4060
2077	PUBLISHED	IA29	9*12*73 -	09*25*73	4/140A,B	(36)	ARC 6X6-FT SUPERSONIC	630
2078	PUBLISHED	IA10	8* 1*73 -	08*03*73	3/139B	(32)	ARC 3.5-FT HYPERSONIC	169
2079	PUBLISHED	LA15	8* 3*73 -	09*24*73	089B, 139 NOSE	(0)	LARC 20-IN HYPERSONIC (M=6)	6441
2080	PUBLISHED	OA57B	9*15*73 -	09*17*73	2A/089B	(2)	RI 7X11-FT LOW SPEED	713
2081	PUBLISHED	OA69	8*28*73 -	09*01*73	3/139B	(43)	RI 7X11-FT LOW SPEED	711
2082	PUBLISHED	OA73	7*11*73 -	7*18*73	3/139B	(42)	ARC 3.5-FT HYPERSONIC	167
2083	PUBLISHED	OA20A	9*10*73 -	09*13*73	4/140A,B	(49)	LARC UNITARY PLAN	1057
2084	PUBLISHED	IA14A	9* 4*73 -	09*13*73	4/140A,B	(47)	ARC 11-FT TRANSONIC	716
2085	PUBLISHED	IH2	9* 4*73 -	09*11*73	3/139	(26)	ARC 3.5-FT HYPERSONIC	171
2086	PUBLISHED	OA71C	9* 4*73 -	09*14*73	3/139B	(43)	RI 7X11-FT LOW SPEED	712
2087	PUBLISHED	SA10F	9*13*73 -	10*01*73	SRB	(449)	MSFC 14-IN TRANSONIC	578
2088	PUBLISHED	SA2FB	9*24*73 -	09*28*73	SRB	(454)	LARC 8-FT TRANSONIC PRESSURE	662
2089	PUBLISHED	OA25	9*14*73 -	09*21*73	4/140A,B	(49)	LARC 8-FT TRANSONIC PRESSURE	661
2090	PUBLISHED	LA8D	7*10*73 -	07*13*73	089B, 139 NOSE	(0)	LARC UNITARY PLAN	1040
2091								
2092	PUBLISHED	OA72	7*30*73 -	08*24*73	3A/139B	(34)	LARC 22-IN HELIUM	415
2093	PUBLISHED	IA37B	10*15*73 -	10*16*73	3A/139B	(34)	MSFC 14-IN TRANSONIC	585
2094	PUBLISHED	OS1	8* 6*73 -	08*10*73	2A/089B	(23)	LARC 26-IN TRANSONIC BLOWDOWN	545
2095	PUBLISHED	OA49	10*18*73 -	11*09*73	4/140A,B	(34)	MSFC 14-IN TRANSONIC	581

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2096	PUBLISHED	OH13	6*13*73	06*13*73	2A/089B	(41)	LARC MACH 8 VARIABLE DENSITY	644
2097	PUBLISHED	OA62A	10* 5*73	10*23*73	4/140A,B	(43)	RI 7X11-FT LOW SPEED	715
2098	PUBLISHED	IH15	8*13*73	08*17*73	2A/089B	(41)	ARC 3.5-FT HYPERSONIC	172
2099	PUBLISHED	OH4B	9*29*73	10*04*73	3/139	(22)	AEDC B / HYPERSONIC	352
2100	PUBLISHED	OH3B	7* 9*73	07*11*73	3/139B	(21)	AEDC B / HYPERSONIC	289
2101	PUBLISHED	OH42C	6*14*73	06*15*73	3/139A, W/CAN	(46)	LARC MACH 8 VARIABLE DENSITY	4080
2102	PUBLISHED	IA15	10*10*73	10*16*73	3/139B	(32)	ARC 3.5-FT HYPERSONIC	175
2103	PUBLISHED	IA62F	11*15*73	11*19*73	4/140A,B	(34)	MSFC 14-IN TRANSONIC	589
2104	PUBLISHED	OA62B	11*13*73	12*06*73	4/140A,B	(43)	RI 7X11-FT LOW SPEED	717
2105	PUBLISHED	IH17	10* 9*73	10*16*73	2A/089B	(41)	LARC MACH 8 VARIABLE DENSITY	646
2106	PUBLISHED	LA14D	12* 5*73	12*07*73	089B, 139 NOSE	(202)	LARC UNITARY PLAN	1058
2107	PUBLISHED	LA20C	8*29*73	08*31*73	089B,139NOSE	(202)	LARC 8-FT TRANSONIC PRESSURE	658
2108	PUBLISHED	IA35	11* 1*73	11*02*73	4/140A,B	(36)	LARC UNITARY PLAN	1063
2109	PUBLISHED	OH45	11* 2*73	11*09*73	3A/139B	(50)	LARC 20-IN FREON	121
2110	PUBLISHED	IH18	10*19*73	10*30*73	2A/089B	(41)	LARC 20-IN FREON	118
2111	PUBLISHED	SA26FB	1*28*74	01*30*74	SRB	(449)	MSFC 14-IN TRANSONIC	595
2112	PUBLISHED	IA57	11*20*73	11*20*73	3/139,089B	(32)	AEDC A / SUPERSONIC	422
2113	PUBLISHED	OA85	10*31*73	11*08*73	3/139B	(32)	LARC 31-IN CONT-FLOW HYP.	101
2114	PUBLISHED	OA86	10*26*73	11*09*73	4/140A,B	(43)	RI 7X11-FT LOW SPEED	716
2115	PUBLISHED	OA87	10*15*73	10*23*73	4/140A,B	(49)	ARC 3.5-FT HYPERSONIC	176
2116	PUBLISHED	OA91	10*26*73	11*01*73	4/140A/B	(49)	RI 7-FT TRISONIC	278
2117	PUBLISHED	OH14	10*17*73	10*18*73	3A/139B	(50)	LARC MACH 8 VARIABLE DENSITY	648
2118	PUBLISHED	IA41	12*11*73	12*14*73	4/140A,B	(67)	LARC 8-FT TRANSONIC PRESSURE	667
2119	PUBLISHED	IA42B	12*17*73	12*21*73	4/140A,B	(67)	LARC UNITARY PLAN	1073
2120	PUBLISHED	OA106	12*17*73	12*18*73	4/140A,B	(67)	LARC 8-FT TRANSONIC PRESSURE	668

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2121	CANCEL	LA38A	12*14*73 -	12*21*73	140A,B	(0)	LARC 8-FT TRANSONIC PRESSURE	669
2122	PUBLISHED	IA69	1*10*74 -	01*14*75	4/140A,B	(67)	RI 7-FT TRISONIC	280
2123	PUBLISHED	IA53	12*20*73 -	01*04*74	2A/089B	(13)	MSFC 14-IN TRANSONIC	588
2124	PUBLISHED	IA16	11*17*73 -	12*04*73	4/140A,B	(36)	ARC 3.5-FT HYPERSONIC	180
2125	PUBLISHED	OA88	12*11*73 -	12*28*73	4/140A,B	(34)	LARC 22-IN HELIUM	7422
2126	CANCEL	LA25	8*30*73 -	09*07*73	3/139B	(32)	LARC 31-IN CONT-FLOW HYP.	100
2127	PUBLISHED	LA35	11*12*73 -	11*13*73	3/139B	(32)	LARC 31-IN CONT-FLOW HYP.	102
2128	PUBLISHED	OA53A	11*19*73 -	11*27*73	4/140A,B	(47)	ARC 11-FT TRANSONIC	747
2129	PUBLISHED	IA14B	9*14*73 -	09*19*73	4/140A,B	(47)	ARC 9X7-FT SUPERSONIC	716
2130	PUBLISHED	OA22A	9*12*73 -	09*14*73	4/140A,B	(47)	ARC 11-FT TRANSONIC	716
2131	PUBLISHED	OA22B	9*19*73 -	09*20*73	4/140A,B	(47)	ARC 9X7-FT SUPERSONIC	716
2132	PUBLISHED	LA42B	7*27*74 -	07*27*74	089B	(0)	AEDC B / HYPERSONIC	48A
2133	PUBLISHED	IA58	2*11*74 -	02*13*74	3/139,089B	(32)	LARC 31-IN CONT-FLOW HYP.	107
2134	PUBLISHED	OA78	12* 3*73 -	12*04*73	4/140A,B	(49)	AEDC C / HYPERSONIC	474
2135	CANCEL	LA13C	11*14*73 -	11*16*73	089B, 139 NOSE	(0)	LARC 31-IN CONT-FLOW HYP.	099
2136	PUBLISHED	IH3	10*31*73 -	11*09*73	3/139	(22)	ARC 3.5-FT HYPERSONIC	178
2137	PUBLISHED	IA60	2*14*74 -	02*20*74	3/139,089B	(32)	LARC 31-IN CONT-FLOW HYP.	108
2138	PUBLISHED	IH4	11*12*73 -	11*16*73	3/139	(26)	LARC UNITARY PLAN	1059
2139	PUBLISHED	OA118	4*24*74 -	04*26*74	4/140A,B	(43)	RI 7X11-FT LOW SPEED	724
2140	PUBLISHED	OA37	1* 7*74 -	01*25*74	4/140A,B	(47)	RI 7X11-FT LOW SPEED	719
2141	PUBLISHED	OH11	10*24*73 -	11*01*73	3/139	(29)	AEDC F / HYPERSONIC	VA35
2142	PUBLISHED	FA4	1*18*74 -	04*15*74	TITAN-3C	(459)	MSFC 14-IN TRANSONIC	587
2143	PUBLISHED	IA61A	1*30*74 -	01*31*74	3/139,089B	(32)	AEDC A / SUPERSONIC	422
2144	PUBLISHED	IA68	1*18*74 -	01*29*74	2A/089B	(13)	RI 7-FT TRISONIC	281
2145	PUBLISHED	TA1F	2*19*74 -	03*05*74	ET	(459)	MSFC 14-IN TRANSONIC	583

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2146	PUBLISHED	IS4	10*18*73 -	10*24*73	2A/089B	(30)	LARC 26-IN TRANSONIC BLOWDOWN	547
2147	PUBLISHED	OA20C	11* 5*73 -	11*08*73	4/140A,B	(49)	LARC UNITARY PLAN	1057
2148	PUBLISHED	IH20	1*18*74 -	02*06*74	3/139	(22)	ARC 3.5-FT HYPERSONIC	185
2149	PUBLISHED	OA90	3* 4*74 -	03*06*74	4/140A,B	(72)	LARC 31-IN CONT-FLOW HYP.	110
2150	PUBLISHED	SA25F	3* 4*74 -	03*11*74	SRB	(454)	LARC UNITARY PLAN	1087
2151	PUBLISHED	OH6	2* 6*74 -	02*11*74	3/139	(22)	ARC 3.5-FT HYPERSONIC	183
2152	PUBLISHED	OA81	11*28*73 -	12*28*73	4/140A,B	(51)	AEDC F / HYPERSONIC	489
2153	PUBLISHED	IH1	12* 3*73 -	12*14*73	3/139	(22)	LARC UNITARY PLAN	1071
2154	PUBLISHED	OH4A	11*12*73 -	12*05*73	3/139	(29)	AEDC B / HYPERSONIC	352
2155	PUBLISHED	OA110	3*15*74 -	03*20*74	4/140A,B	(16)	RI 7X11-FT LOW SPEED	721
2156	PUBLISHED	IA17A	3* 6*74 -	03*15*74	3/139B	(52)	AEDC B / HYPERSONIC	422
2157	PUBLISHED	IH19B	12*27*73 -	01*08*74	2A/089B	(50)	LARC HYPERSONIC NITROGEN	28
2158	PUBLISHED	IS6B	3*20*73 -	05*27*73	2A/089B	(13)	MSFC 14-IN TRANSONIC	559
2159	PUBLISHED	OA59	3*13*74 -	3*21*74	4/140A,B	(49)	ARC 6X6-FT SUPERSONIC	709
2160	PUBLISHED	IA18	4* 9*74 -	04*12*74	3/139B	(52)	ARC 3.5-FT HYPERSONIC	191
2161	PUBLISHED	SA6F	12* 3*73 -	01*16*74	SRB	(454)	LERC 10X10-FT SUPERSONIC	035
2162	PUBLISHED	OA36	2*25*74 -	03*01*74	4/140A,B	(49)	ARC 3.5-FT HYPERSONIC	187
2163	PUBLISHED	OA20B	4* 8*74 -	04*12*74	4/140A,B	(49)	LARC UNITARY PLAN	1097
2164	PUBLISHED	IH21	10*29*73 -	12*13*73	3/139	(37)	CALSPAN HYPERSONIC SHOCK	100
2165	PUBLISHED	TA2F	4*29*74 -	09*23*74	ET	(460)	MSFC 14-IN TRANSONIC	596
2166	PUBLISHED	IH1C	7* 6*73 -	07*13*73	2A/089B	(41)	LARC UNITARY PLAN	1041
2167	PUBLISHED	OA98	3*27*74 -	04*03*74	4/140A/B	(49)	ARC 3.5-FT HYPERSONIC	190
2168	PUBLISHED	LA32B	11*28*73 -	12*03*73	F.S. TILE ARRAY	(0)	LARC 31-IN CONT-FLOW HYP.	097
2169	PUBLISHED	IA81A	7*26*74 -	08*27*74	4/140A,B (MOD)	(47)	ARC 11-FT TRANSONIC	019
2170	PUBLISHED	IA19A	9*16*74 -	09*23*74	5/140C	(88)	ARC 11-FT TRANSONIC	014

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2171	PUBLISHED	OH38	6*21*74 -	07*18*74	4/140B	(61)	ARC 3.5-FT HYPERSONIC	198
2172	PUBLISHED	OA99	3*26*74 -	4*12*74	3/139B	(21)	LARC 60-FT. VACUUM SPHERE	3289
2173	PUBLISHED	IA8	2*12*73 -	03*12*73	ATP	(6)	ARC 14-FT TRANSONIC	711
2174	PUBLISHED	IA33	5* 9*74 -	07*21*74	5/140C	(74)	MSFC 14-IN TRANSONIC	594
2175	PUBLISHED	IA70	5* 3*74 -	05*24*74	4/140A,B	(49)	RI 7-FT TRISONIC	282
2176	PUBLISHED	LA40	5*13*74 -	06*07*74	139B	(0)	LARC 22-IN HELIUM	7426
2177	PUBLISHED	OA83	5* 8*74 -	05*16*74	4/140A,B	(36)	ARC 3.5-FT HYPERSONIC	194
2178	PUBLISHED	OA53B	11*12*73 -	11*16*73	4/140A,B	(47)	ARC 9X7-FT SUPERSONIC	747
2179	PUBLISHED	OS8B	7*19*74 -	07*29*74	HRSI TILE	(81)	ARC 9X7-FT SUPERSONIC	705
2180	PUBLISHED	IH28-2	5*20*74 -	05*24*74	2A/089B	(50)	ARC 3.5-FT HYPERSONIC	195
2181	PUBLISHED	TA9F	6* 3*74 -	06*15*74	ET	(466)	ARC 3.5-FT HYPERSONIC	196
2182	PUBLISHED	LA49B	7*15*74 -	07*17*74	089B-MOD NOSE	(0)	LARC UNITARY PLAN	1111
2183	PUBLISHED	LA51	5*24*74 -	05*31*74	140A,B	(0)	LARC 8-FT TRANSONIC PRESSURE	684
2184	PUBLISHED	LA48	4*10*74 -	04*15*74	089B-MOD NOSE	(0)	LARC 8-FT TRANSONIC PRESSURE	680
2185	PUBLISHED	OA53C	11*28*73 -	12*06*73	4/140A,B	(47)	ARC 8X7-FT SUPERSONIC	747
2186	PUBLISHED	OA116	6*10*74 -	06*14*74	4/140A,B	(49)	LARC 8-FT TRANSONIC PRESSURE	686
2187	PUBLISHED	OA119A	6*17*74 -	06*25*74	4/140A,B	(16)	RI 7X11-FT LOW SPEED	726
2188	PUBLISHED	LA39C	4* 1*74 -	04*08*74	140A,B	(0)	LARC UNITARY PLAN	1075
2189	PUBLISHED	IA110-2	7* 8*74 -	07*11*74	4/140A,B	(67)	ARC 9X7-FT SUPERSONIC	052
2190	PUBLISHED	OA108	6*24*74 -	07*09*74	5/140C	(74)	MSFC 14-IN TRANSONIC	599
2191	PUBLISHED	LA47C	7* 8*74 -	07*10*74	140A/B ORB	(0)	LARC 31-IN CONT-FLOW HYP.	104
2192	PUBLISHED	IA87	7*18*74 -	07*20*74	3/139B	(52)	AEDC A / SUPERSONIC	60A
2193	PUBLISHED	OH26	7*22*74 -	07*29*74	4/140B	(22)	ARC 3.5-FT HYPERSONIC	199
2194	PUBLISHED	IA81B	8* 9*74 -	08*22*74	4/140A,B (MOD)	(47)	ARC 9X7-FT SUPERSONIC	019
2195	PUBLISHED	OA82	8*12*74 -	08*16*74	4/140A,B	(32)	LARC 31-IN CONT-FLOW HYP.	113

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2196	PUBLISHED	OA79	8* 1*74 -	08*03*74	4/140A,B(MOD)	(49)	AEDC B / HYPERSONIC	71A
2197	PUBLISHED	FH10	1*21*74 -	01*29*74	3/139	(22)	AEDC F / HYPERSONIC	291
2198	PUBLISHED	OA115A	7*29*74 -	07*31*74	4/140A,B(MOD)	(49)	AEDC A / SUPERSONIC	71A
2199	PUBLISHED	LA43B	3*18*74 -	03*27*74	4/140A,B	(0)	LARC UNITARY PLAN	1093
2200	PUBLISHED	LA44	4* 2*74 -	04*09*74	4/140A,B	(0)	LARC 8-FT TRANSONIC PRESSURE	677
2201	PUBLISHED	CA3	8*15*74 -	08*30*74	4/140A,B/747	(43)	UNIV. OF WASH. LOW SPEED	1136
2202	PUBLISHED	OA123	9* 6*74 -	09*10*74	4/140A,B (ALT)	(43)	RI 7X11-FT LOW SPEED	731
2203	PUBLISHED	OA119B	8*22*74 -	09*06*74	4/140A,B	(16)	RI 7X11-FT LOW SPEED	730
2204	PUBLISHED	IA43	8*26*74 -	09*03*74	4/140A,B	(72)	LARC 8-FT TRANSONIC PRESSURE	693
2205	PUBLISHED	OA109	8*26*74 -	08*29*74	5/140C	(74)	LARC 22-IN HELIUM	431
2206	PUBLISHED	IA44B	8*19*74 -	08*23*74	4/140A,B	(72)	LARC UNITARY PLAN	1119
2207	PUBLISHED	SA29F	8* 8*74 -	09*18*74	SRB FORE BODY	(467)	CALSPAN 32-IN LUDWIEG	033
2208	PUBLISHED	TA3F	9*27*74 -	10*11*74	ET	(470)	MSFC 14-IN TRANSONIC	609
2209	PUBLISHED	OA124	10*14*74 -	10*23*74	4/140A,B	(43)	RI 7X11-FT LOW SPEED	736
2210	PUBLISHED	IH27	9* 7*74 -	09*25*74	TPS TILES	(15)	ARC 3.5-FT HYPERSONIC	200
2211	PUBLISHED	CA5	9*20*74 -	09*30*74	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1431
2212	PUBLISHED	IA80	11* 4*74 -	11*08*74	5/140C	(88)	ARC 11-FT TRANSONIC	023
2213	IN PROCESS	LA54	8*14*74 -	08*19*74	140C ORB.	(0)	LARC 20-IN HYPERSONIC (M=6)	6456
2214	PUBLISHED	OA89	7*15*74 -	08*05*74	5/140C	(74)	LARC HYPERSONIC NITROGEN	30
2215	PUBLISHED	LA58	9*30*74 -	10*04*74	140A,B	(42)	LTV 4X4-FT SUPERSONIC	512
2216	PUBLISHED	SH12F	7*29*74 -	08*07*74	SRB	(0)	LARC UNITARY PLAN	1115
2217	PUBLISHED	CA20	10* 9*74 -	10*15*74	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1431
2218	PUBLISHED	TH1F	9* [illegible]*74 -	09*09*74	ET	(0)	AEDC F / HYPERSONIC	25A
2219	PUBLISHED	IA82C	11*11*74 -	11*15*74	5/140C	(75)	ARC 8X7-FT SUPERSONIC	044
2220	PUBLISHED	LA52	8*26*74 -	08*30*74	140A,B	(0)	LARC 20-IN HYPERSONIC (M=6)	6458

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2221	PUBLISHED	OA143	11* 6*74 -	11*11*74	4/140A,B	(16)	RI 7X11-FT LOW SPEED	737
2222	PUBLISHED	OH49B	7* 2*74 -	07*12*74	4/140B	(22)	AEDC B / HYPERSONIC	57A
2223	PUBLISHED	SA8F	10*18*74 -	12*10*74	SRB	(471)	MSFC 14-IN TRANSONIC	604
2224	PUBLISHED	LA56	11*11*74 -	11*22*74	VEH. 5	(0)	LARC 8-FT TRANSONIC PRESSURE	699
2225	PUBLISHED	OH4C	9*26*73 -	09*26*73	3/139B	(21)	AEDC B / HYPERSONIC	352
2226	PUBLISHED	IA61B	2*26*74 -	02*26*74	3/139,089B	(52)	AEDC A / SUPERSONIC	21AA
2227	PUBLISHED	IA71B-2	12*19*74 -	01*09*75	5/140C	(74)	MSFC 14-IN TRANSONIC	610
2228	PUBLISHED	LA46B	9*24*74 -	10*10*74	140A,B ORB	(0)	LARC UNITARY PLAN	1117
2229	PUBLISHED	OA102	6*17*74 -	06*18*74	4/140A,B	(36)	LARC 8-FT TRANSONIC PRESSURE	687
2230	PUBLISHED	IA17B	3*18*74 -	03*19*74	3/139B	(52)	AEDC B / HYPERSONIC	422
2231	PUBLISHED	IA82B	1*28*75 -	02*04*75	5/140C	(75)	ARC 9X7-FT SUPERSONIC	044
2232	PUBLISHED	OA131	9*11*74 -	09*26*74	5/140C	(74)	MSFC 14-IN TRANSONIC	607
2233	PUBLISHED	LA59	12*20*74 -	01*07*75	4/140A,B	(72)	LARC 8-FT TRANSONIC PRESSURE	703
2234	PUBLISHED	OA113	8*10*74 -	10*04*74	4/140A,B	(51)	CALSPAN HYPERSONIC SHOCK	184-
2235	PUBLISHED	SA30F	3* 3*75 -	03*13*75	SRB	(473)	MSFC 14-IN TRANSONIC	611
2236	PUBLISHED	CA11	2*12*75 -	02*20*75	ET/747	(0)	UNIV. OF WASH. LOW SPEED	1146
2237	IN PROCESS	OA155	2*10*75 -	03*07*75	4/140A,B (MOD)	(47)	LARC V/STOL	114
2238	PUBLISHED	OA93	11*18*74 -	11*23*74	4/140A,B	(51)	CALSPAN HYPERSONIC SHOCK	737
2239	PUBLISHED	LA38B	3*27*74 -	03*29*74	140A,B	(0)	LARC 8-FT TRANSONIC PRESSURE	676
2240	PUBLISHED	IH41A	3*31*75 -	5*21*75	5/140C	(60)	AEDC A / SUPERSONIC	4A
2241	PUBLISHED	OH39B	1* 8*75 -	01*09*75	5/140C	(60)	AEDC B / HYPERSONIC	74A
2242	PUBLISHED	IA111	3*21*75 -	03*28*75	3/139B	(52)	AEDC A / SUPERSONIC	A3A
2243	PUBLISHED	CA23A	3*21*75 -	04*17*75	140C(MOD)/747	(48)	ARC 14-FT TRANSONIC	085
2244	PUBLISHED	SA28F-2	3*17*75 -	04*11*75	SRB	(469)	MSFC 14-IN TRANSONIC	603
2245	PUBLISHED	OA161C	3*26*75 -	03*31*75	140A,B (MOD)	(45)	ARC 8X7-FT SUPERSONIC	094

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2246								
2247	PUBLISHED	OA160	2* 5*75	- 02*08*75	4/140A,B	(51)	AEDC F / HYPERSONIC	28A
2248	PUBLISHED	IH48	4*17*75	- 05*08*75	5/140C	(60)	ARC 3.5-FT HYPERSONIC	211
2249	PUBLISHED	IH33B	12* 5*74	- 12*19*74	5/140C	(37)	CALSPAN HYPERSONIC SHOCK	131
2250	PUBLISHED	OH43	12* 2*73	- 12*21*73	TPS TILES	(15)	ARC 3.5-FT HYPERSONIC	182
2251	PUBLISHED	OH9	9*13*73	- 09*21*73	3/139	(29)	AEDC B / HYPERSONIC	353
2252	PUBLISHED	OH25A	8*21*74	- 08*22*74	3/139B	(21)	AEDC B / HYPERSONIC	83A
2253	PUBLISHED	IA125-2	4*25*75	- 05*22*75	5/140C(74TS)	(77)	MSFC 14-IN TRANSONIC	622
2254	PUBLISHED	OA148	5* 5*75	- 05*17*75	4/140A,B (MOD)	(47)	ARC 11-FT TRANSONIC	073
2255	PUBLISHED	AA3B	2* 1*71	- 02*15*71	PRE-ATP	(0)	ARC 9X7-FT SUPERSONIC	608
2256	IN PROCESS	LA68	2*26*75	- 03*20*75	140C ORB	(0)	LARC 22-IN HELIUM	439
2257	PUBLISHED	LA69	4*24*75	- 04*29*75	5/140C	(72)	LARC 8-FT TRANSONIC PRESSURE	714
2258	PUBLISHED	IA72	5*19*75	- 05*31*75	5/140C	(88)	ARC 11-FT TRANSONIC	072
2259								
2260								
2261	PUBLISHED	OA100	5*27*75	- 06*14*75	OV101(ALT)	(76)	ARC 40X80-FT SUBSONIC	462
2262	PUBLISHED	CA6	5*20*75	- 06*06*75	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1472
2263	PUBLISHED	OH74	6* 3*75	- 06*12*75	5/140C	(56)	AEDC B / HYPERSONIC	87A
2264	PUBLISHED	LA62	5*14*75	- 05*23*75	140C/REMOTE ELE	(44)	LARC 8-FT TRANSONIC PRESSURE	717
2265	PUBLISHED	OA159	6*23*75	- 07*08*75	140A,B/(ALT)	(45)	ARC 12-FT PRESSURE	078
2266	PUBLISHED	LA67	6*20*75	- 07*02*75	140C/REMOTE ELE	(44)	LTV 4X4-FT SUPERSONIC	552
2267	PUBLISHED	MA22	5* 6*75	- 06*03*75	4/140A,B	(32)	LARC 31-IN CONT-FLOW HYP.	118
2268	PUBLISHED	CA9	6*25*75	- 07*14*75	4/140A,B/747	(47)	THE BOEING CO. - TRANSONIC	1477
2269	PUBLISHED	LA70	7*28*75	- 08*06*75	140C/REMOTE ELE	(44)	CALSPAN 8-FT TRANSONIC	103
2270	PUBLISHED	LA63A	7*18*75	- 07*18*75	140C/REMOTE ELE	(44)	LARC UNITARY PLAN	1118

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED. - COMPL.	MODEL REF.	(ID)	WIND TUNNEL FACILITY	NO.
2271	PUBLISHED	LA71A	10*17*75 - 10*22*75	4/140A,B	(69)	LARC UNITARY PLAN	1132
2272	PUBLISHED	IA114	8*18*75 - 08*22*75	5/140C	(52)	AEDC B / HYPERSONIC	C4A
2273	PUBLISHED	CA26	8* 4*75 - 08*15*75	140C(MOD)/747	(48)	LTV 4X4-FT SUPERSONIC	559
2274	PUBLISHED	FA14	1* 9*75 - 07*06*75	5/140C	(74)	MSFC 14-IN TRANSONIC	600
2275	PUBLISHED	CA23B	5* 1*75 - 07*22*75	140C(MOD)/747	(48)	ARC 14-FT TRANSONIC	085
2276	PUBLISHED	FH13	9*22*75 - 09*25*75	ET/SPIKE	(0)	AEDC A / SUPERSONIC	E1A
2277	PUBLISHED	SA13F	9*30*74 - 06*17*75	SRB	(461)	MSFC 32-IN LUDWIEG (HIGH RN)	034
2278	CANCEL	LA61A	8*25*75 - 09*10*75	140C/REMOTE ELE	(44)	LARC LOW TURBULANCE PRESSURE	219
2279	PUBLISHED	LA63B	9*12*75 - 09*17*75	140C/REMOTE ELE	(44)	LARC UNITARY PLAN	1151
2280	PUBLISHED	LA28	6*17*74 - 06*20*74	140A,B ORB	(0)	LTV 4X4-FT SUPERSONIC	498
2281	PUBLISHED	LA66	10*20*75 - 10*24*75	140C/REMOTE ELE	(44)	ARC 12-FT PRESSURE	135
2282	PUBLISHED	IH34	5* 5*75 - 09*03*75	5/140C	(19)	LERC 10X10-FT SUPERSONIC	038
2283	PUBLISHED	MA14	4*23*73 - 05*02*73	2A/089B(CAN)	(95)	LTV 15X20-FT LOW SPEED	422
2284	PUBLISHED	IS2B	9*25*75 - 10*29*75	5/140C	(84)	ARC 9X7-FT SUPERSONIC	113
2285	PUBLISHED	OH50A	3*29*74 - 04*11*74	5/140C	(82)	AEDC B / HYPERSONIC	526
2286	PUBLISHED	OA220	11*11*75 - 11*21*75	VEH 101 (ADS)	(57)	ARC 14-FT TRANSONIC	150
2287	IN PROCESS	OS13	11*24*75 - 11*26*75	LRSI TILE	(85)	ARC 9X7-FT SUPERSONIC	166
2288	PUBLISHED	OH64	4*14*75 - 06*20*75	2A/089B	(25)	LERC SPACE POWER FACILITY	OH64
2289	PUBLISHED	OA163A	11*24*75 - 12*09*75	4/140A,B	(16)	RI 7X11-FT LOW SPEED	751
2290	PUBLISHED	CA8	8*18*75 - 09*12*75	4/140A,B/747	(43)	LARC V/STOL	129
2291	IN PROCESS	LA79	11*28*75 - 12*11*75	140C	(0)	NSWC HYPERVELOCITY LAB (#8A)	1275
2292	PUBLISHED	LA36B	6* 3*75 - 06*05*75	140A,B	(32)	LARC LOW TURBULANCE PRESSURE	214
2293	PUBLISHED	IA40	6*23*76 - 06*29*76	5/140C	(75)	AEDC A / SUPERSONIC	425
2294	PUBLISHED	OA172	12*15*75 - 01*13*76	4/140A,B([illegible])	(43)	RI 7X11-FT LOW SPEED	752
2295	PUBLISHED	IH41B	12*11*75 - 01*09*76	5/140C	(60)	AEDC A / SUPERSONIC	4A

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED. - COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2296	PUBLISHED	LA81	1*14*76 - 01*23*76	ORB/TC (ALT)	(0)	LARC LOW TURBULANCE PRESSURE	229
2297							
2298	PUBLISHED	LA73B	12*10*76 - 12*13*75	4/140A,B	(69)	LARC LOW TURBULANCE PRESSURE	238
2299	PUBLISHED	LA80	10* 6*75 - 11*07*75	140C/747	(88)	LARC 7X10-FT HIGH SPEED	999
2300	PUBLISHED	LA61B	1* 5*76 - 01*14*76	140C/REMOTE ELE	(44)	LARC LOW TURBULANCE PRESSURE	228
2301	PUBLISHED	OH54A	10* 4*74 - 10* 8*74	5/140C	(82)	AEDC B / HYPERSONIC	82A
2302	PUBLISHED	OA174	2* 2*76 - 02*27*76	OV101(ALT)	(76)	ARC 40X80-FT SUBSONIC	479
2303	PUBLISHED	OH75	9* 2*75 - 09*03*75	5/140C	(82)	AEDC B / HYPERSONIC	E3A
2304	PUBLISHED	OA173	3*15*76 - 03*26*76	140C(ALT)	(45)	ARC 12-FT PRESSURE	180
2305	PUBLISHED	LA76	2*25*76 - 03*06*76	140C/REMOTE ELE	(44)	LTV 4X4-FT SUPERSONIC	573
2306	PUBLISHED	IA135C	3*12*76 - 03*23*76	4/140A,B (MOD)	(47)	ARC 8X7-FT SUPERSONIC	144
2307	PUBLISHED	CA14	11*13*75 - 12*02*75	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1496
2308	PUBLISHED	IH5	1*21*74 - 07*22*74	2A/089B	(19)	CALSPAN 32-IN LUDWIEG	181
2309	PUBLISHED	LA72	3*26*76 - 03*31*76	4/140A,B	(69)	LARC 8-FT TRANSONIC PRESSURE	740
2310	PUBLISHED	SA14FB	1* 6*76 - 01*11*76	SRB	(486)	MSFC 14-IN TRANSONIC	640
2311	PUBLISHED	LA88	5*21*75 - 05*2?*75	089B	(13)	LARC 20-IN HYPERSONIC (M=6)	6468
2312	PUBLISHED	IH47	3* 8*76 - 03*19*76	5/140C	(60)	AEDC A / SUPERSONIC	J3A
2313	PUBLISHED	FH14	3*15*76 - 04*06*76	ET/SPIKE	(0)	ARC 3.5-FT HYPERSONIC	215
2314	PUBLISHED	OA176	3*29*76 - 04*15*76	4/140A,B(ALT)	(43)	RI 7X11-FT LOW SPEED	754
2315	PUBLISHED	IA141	3*31*76 - 04*05*76	5/140C	(72)	RI 7-FT TRISONIC	297
2316	PUBLISHED	IA137	4*26*76 - 05*03*76	ET FORETANK	(68)	ARC 14-FT TRANSONIC	143
2317	PUBLISHED	OH53B	4*14*76 - 04*23*76	5/140C	(82)	ARC 3.5-FT HYPERSONIC	216
2318	PUBLISHED	LA75	4* 6*76 - 04*16*76	140C/REMOTE ELE	(44)	LARC UNITARY PLAN	1173
2319	PUBLISHED	IH43	12*17*75 - 02*23*76	5/140C	(59)	CALSPAN HYPERSONIC SHOCK	189
2320	PUBLISHED	OA169	3*26*76 - 04*09*76	5/140C	(70)	AEDC B / HYPERSONIC	D8A

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2321	PUBLISHED	OH69	11*14*75 -	12*11*75	5/140C	(82)	AEDC B / HYPERSONIC	E9A
2322	PUBLISHED	OA228	5*29*76 -	05*01*76	VEH 102 (ADS)	(57)	RI 7X11-FT LOW SPEED	757
2323	PUBLISHED	IA94A	4*18*76 -	04*23*76	5/140C	(72)	LARC UNITARY PLAN	1152
2324	PUBLISHED	IA94B	4*26*76 -	05*04*76	5/140C	(72)	LARC UNITARY PLAN	1177
2325	PUBLISHED	SA14FA	12*23*75 -	03*19*76	SRB	(449)	MSFC 14-IN TRANSONIC	620
2326	PUBLISHED	IA93	5*10*76 -	05*14*76	5/140C	(72)	LARC 8-FT TRANSONIC PRESSURE	749
2327	PUBLISHED	IA22	5* 3*76 -	05*08*76	5/140C	(70)	AEDC B / HYPERSONIC	59A
2328	PUBLISHED	LA34	1*17*74 -	01*31*74	F.S. TILE ARRAY	(0)	LARC 31-IN CONT-FLOW HYP.	105
2329	PUBLISHED	OA224	2*23*76 -	03*24*76	VEH 102 (ADS)	(57)	LARC 16-FT TRANSONIC	312
2330	PUBLISHED	OH52	5* 6*74 -	05*15*74	3/139B	(29)	AEDC B / HYPERSONIC	524
2331	PUBLISHED	SA11FC	3*29*76 -	04*14*76	SRB	(483)	ARC 8X7-FT SUPERSONIC	074
2332	PUBLISHED	CA13	6* 8*76 -	07*01*76	140C(ALT)/747	(45)	ARC 14-FT TRANSONIC	121
2333	PUBLISHED	OA175	6*28*76 -	07*09*76	140A,B (ALT)	(47)	ARC 11-FT TRANSONIC	187
2334	PUBLISHED	SA16F	5* 5*76 -	05*06*76	SRB	(486)	AEDC 4-FT TRANSONIC	445
2335	PUBLISHED	IA140B	10* 1*76 -	01*28*77	5/140C	(74)	MSFC 14-IN TRANSONIC	646
2336	PUBLISHED	LA145B	9*11*81 -	09*17*81	140C	(203)	LARC UNITARY PLAN	1345
2337	PUBLISHED	OA236	5*28*76 -	06*02*76	ADS PROBES	(99)	RI 7X11-FT LOW SPEED	759
2338	PUBLISHED	CS3	9*12*75 -	09*15*75	140A,B/747	(8)	UNIV. OF WASH. LOW SPEED	1170
2339	IN PROCESS	OS32	7*15*76 -	07*27*76	TILE PANEL	(35)	ARC 2X2-FT TRANSONIC	167
2340	PUBLISHED	OH98B	7*26*76 -	07*26*76	5/140C	(60)	AEDC B / HYPERSONIC	074
2341	PUBLISHED	CS5	11* 3*75 -	11*05*75	140A,B/747	(45)	THE BOEING CO. - TRANSONIC	1493
2342	PUBLISHED	OH54C	8*26*75 -	09*02*75	5/140C	(82)	AEDC B / HYPERSONIC	82A
2343	PUBLISHED	LA85	4* 7*76 -	05*24*76	140C	(13)	LARC 22-IN HELIUM	445
2344	PUBLISHED	LA77	7* 9*76 -	07*24*76	140C/REMOTE ELE	(44)	ARC 11-FT TRANSONIC	200
2345	PUBLISHED	SA21F	9*16*76 -	10*06*76	SRB	(486)	MSFC 14-IN TRANSONIC	645

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2346	PUBLISHED	IA142	8*11*76 -	08*18*76	5/140C	(75)	AEDC A / SUPERSONIC	K1A
2347	PUBLISHED	CA15A	10*16*75 -	11*01*75	4/140A,B/747	(43)	UNIV. OF WASH. LOW SPEED	1173
2348	PUBLISHED	CA15B	11*19*75 -	11*26*75	4/140A,B/747	(43)	UNIV. OF WASH. LOW SPEED	1178
2349	PUBLISHED	CA17	6*21*76 -	07*02*76	4/140A,B/747	(43)	UNIV. OF WASH. LOW SPEED	1184
2350	PUBLISHED	OH46	11*12*73 -	12*07*73	4/140B	(90)	LARC MACH 8 VARIABLE DENSITY	4556
2351	PUBLISHED	OA238	10*25*76 -	11*08*76	ADS PROBES	(99)	RI 7X11-FT LOW SPEED	764
2352	PUBLISHED	LA91	9* 3*76 -	09*15*76	140C/REMOTE ELE	(44)	LARC 8-FT TRANSONIC PRESSURE	758
2353	PUBLISHED	LA89	10* 4*76 -	10*18*76	OV101 (ALT)	(201)	ARC 11-FT TRANSONIC	213
2354	PUBLISHED	IA143	11* 8*76 -	11*13*76	5/140C	(75)	AEDC A / SUPERSONIC	P8A
2355	PUBLISHED	OH49A	4* 3*74 -	04*06*74	3/139B	(22)	AEDC B / HYPERSONIC	525
2356	PUBLISHED	OH60	5*12*75 -	05*12*75	5/140C	(83)	AEDC B / HYPERSONIC	B7A
2357	PUBLISHED	IH68	10*12*76 -	12*08*76	5/140C	(60)	ARC 3.5-FT HYPERSONIC	222
2358	PUBLISHED	OH50B	7*12*74 -	07*17*74	5/140C	(83)	AEDC B / HYPERSONIC	58A
2359	PUBLISHED	OH66	8*30*76 -	10*17*76	5/140C	(66)	CALSPAN HYPERSONIC SHOCK	131
2360	PUBLISHED	OA221C	11*15*76 -	11*22*76	ADS PROBES	(99)	ARC 8X7-FT SUPERSONIC	119
2361	PUBLISHED	OA163B	12*21*76 -	12*23*76	4/140A,B	(16)	RI 7X11-FT LOW SPEED	788
2362	IN PROCESS	LA92	11*11*76 -	11*19*76	OV101	(201)	LARC 8-FT TRANSONIC PRESSURE	764
2363	PUBLISHED	OS7	8*12*74 -	08*30*74	4/140B	(55)	LARC 16-FT TRANSONIC DYNAMIC	246
2364	PUBLISHED	OA145B	4*15*77 -	05*03*77	VEH 102	(39)	ARC 9X7-FT SUPERSONIC	118
2365	PUBLISHED	OS6	9* 2*74 -	09*12*74	4/140B	(54)	LARC 16-FT TRANSONIC DYNAMIC	246
2366	PUBLISHED	OH25B	1*30*75 -	02*03*75	5/140C	(56)	AEDC B / HYPERSONIC	83A
2367	PUBLISHED	OH57B	12* 4*76 -	12*05*76	140C	(92)	AEDC B / HYPERSONIC	K3A
2368	PUBLISHED	OH51-3	6*26*74 -	07*03*74	4/140B	(90)	LARC 31-IN CONT-FLOW HYP.	112
2369	PUBLISHED	SA31F	4*27*76 -	02*01*77	SRB	(487)	MSFC 32-IN LUDWIEG (HIGH RN)	039
2370	PUBLISHED	OA149C	2*16*77 -	02*18*77	5/140C	(47)	ARC 8X7-FT SUPERSONIC	115

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2371	PUBLISHED	OH78	7* 2*76 -	11*24*76	5/140C	(65)	JSC VAC. CHAMBER A	56-A
2372	PUBLISHED	IH72	1* 3*77 -	01*10*77	5/140C	(60)	AEDC A / SUPERSONIC	K2A
2373	PUBLISHED	LA99	2*17*77 -	02*28*77	TAILCONE	(201)	LARC 8-FT TRANSONIC PRESSURE	769
2374	PUBLISHED	LA103	3*25*77 -	04*04*77	SUPPORT TARES	(0)	CALSPAN 8-FT TRANSONIC	113
2375	PUBLISHED	OA237	1*24*77 -	01*31*77	ADS PROBES	(99)	ARC 40X80-FT SUBSONIC	500
2376	PUBLISHED	OA149A	2*24*77 -	03*04*77	5/140C	(47)	ARC 11-FT TRANSONIC	115
2377	PUBLISHED	IA144	4* 6*77 -	04*15*77	5/140C	(72)	ARC 11-FT TRANSONIC	228
2378	PUBLISHED	IA191	6*20*80 -	06*27*80	FUEL LINE	(112)	ARC 11-FT TRANSONIC	412
2379								
2380	PUBLISHED	OA145A	3* 8*77 -	04*02*77	VEH 102	(39)	ARC 11-FT TRANSONIC	118
2381								
2382	PUBLISHED	OH8F	5*15*74 -	07*16*74	2A/089B	(25)	MSFC IMPULSE BASE FLOW FAC.	027
2383	IN PROCESS	LA93	3*16*77 -	04*21*77	NOSE CONE	(0)	LARC 31-IN CONT-FLOW HYP.	130
2384	PUBLISHED	IA148	4*27*77 -	05*03*77	5/140C	(70)	AEDC B / HYPERSONIC	TOA
2385	PUBLISHED	OH15	9*12*73 -	09*20*73	FLAT PLATE	(53)	ARC 3.5-FT HYPERSONIC	173
2386	PUBLISHED	OH44	10*24*73 -	10*30*73	FLAT PLATE	(53)	ARC 3.5-FT HYPERSONIC	177
2387								
2388	PUBLISHED	OH84A-2	4*20*77 -	04*21*77	5/140C	(83)	AEDC B / HYPERSONIC	R4A
2389	PUBLISHED	OA145C	4* 6*77 -	04*20*77	VEH 102	(39)	ARC 8X7-FT SUPERSONIC	118
2390	PUBLISHED	LA101	5*18*77 -	05*24*77	140C/REMOTE ELE	(44)	LARC UNITARY PLAN	1194
2391	PUBLISHED	IA244	5*24*77 -	06*01*77	5/140C	(72)	LARC 8-FT TRANSONIC PRESSURE	779
2392	PUBLISHED	OA250	7* 1*77 -	07*07*77	140C(ALT)	(45)	RI 7X11-FT LOW SPEED	775
2393	PUBLISHED	IH51A	7*14*77 -	07*27*77	FLAT PLATE	(58)	ARC 3.5-FT HYPERSONIC	228
2394								
2395	PUBLISHED	LA111	8* 3*77 -	08*05*77	140C SILTS	(44)	LARC 8-FT TRANSONIC PRESSURE	786

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2396	PUBLISHED	LA110	8* 8*77 -	08*10*77	140C SILTS	(44)	LARC UNITARY PLAN	1212
2397	PUBLISHED	LA113	8* 5*77 -	09*08*77	5/140C	(72)	LARC 8-FT TRANSONIC PRESSURE	787
2398	PUBLISHED	IA105A	9* 2*77 -	11*20*77	5/140C	(47)	AEDC 16-FT TRANSONIC	470
2399	PUBLISHED	LA114	8*23*77 -	08*31*77	140C SILTS	(44)	LARC UNITARY PLAN	1217
2400	PUBLISHED	OA234	6* 7*77 -	08*11*77	ADS PROBES	(99)	LERC 10X10-FT SUPERSONIC	042
2401	PUBLISHED	IS1C	8* 9*73 -	08*11*73	2A/089B	(11)	ARC 8X7-FT SUPERSONIC	705
2402	PUBLISHED	OA223	11*20*76 -	11*30*76	VEH 102	(39)	RI 7X11-FT LOW SPEED	776
2403	PUBLISHED	IA156A	10*28*77 -	11*10*77	VEH 102	(89)	AEDC 16-FT TRANSONIC	470
2404	PUBLISHED	IA119	10* 7*77 -	10*31*77	5/140C	(88)	ARC 11-FT TRANSONIC	275
2405	PUBLISHED	OA101	9*13*77 -	11*11*77	VEH 102	(39)	ARC 12-FT PRESSURE	218
2406	PUBLISHED	IA181	12*15*77 -	02*03*78	5/140C	(74)	MSFC 14-IN TRANSONIC	649
2407	PUBLISHED	IH73	12* 1*77 -	01*23*78	5/140C	(50)	ARC 3.5-FT HYPERSONIC	233
2408	PUBLISHED	IA156B	12*16*77 -	01*06*78	VEH 102	(89)	ARC 9X7-FT SUPERSONIC	272
2409	PUBLISHED	LA115	2* 1*78 -	02*06*78	140C/REMOTE ELE	(44)	LARC 8-FT TRANSONIC PRESSURE	803
2410	PUBLISHED	OH56	12* 6*77 -	12*10*77	WING TIP SEAL	(91)	AEDC B / HYPERSONIC	R3A
2411	CANCEL	LA116	2* 6*78 -	02*06*78	140C	(201)	LARC 8-FT TRANSONIC PRESSURE	804
2412	PUBLISHED	IH90	1*30*78 -	03*10*78	5/140C	(60)	ARC 3.5-FT HYPERSONIC	234
2413	PUBLISHED	IA105B	1* 9*78 -	02*01*78	5/140C	(47)	ARC 9X7-FT SUPERSONIC	242
2414	PUBLISHED	OA232	2*17*78 -	03*01*78	ADS PROBES	(99)	AEDC 16-FT TRANSONIC	431
2415	PUBLISHED	OA209	3*21*78 -	03*30*78	VEH. 102	(105)	AEDC A / SUPERSONIC	P5A
2416	PUBLISHED	OA208	3*30*78 -	04*06*78	VEH. 102	(105)	AEDC B / HYPERSONIC	P5A
2417	PUBLISHED	OH58	3*24*78 -	04*21*78	ELEV/ELEV SEAL	(93)	ARC 3.5-FT HYPERSONIC	235
2418	PUBLISHED	IH100	6*20*77 -	06*23*77	GAS TEMP PROBE	(0)	ARC 3.5-FT HYPERSONIC	227
2419	PUBLISHED	OA270C	4* 8*78 -	04*28*78	VEH 102	(104)	LARC 16-FT TRANSONIC	325
2420	PUBLISHED	OH103A	2*20*78 -	02*21*78	VEH. 5 F'BODY	(83)	AEDC B / HYPERSONIC	V2C

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED. - COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2421	PUBLISHED	OA251C	5*29*78 - 06*15*78	ADS PROBES	(99)	ARC 8X7-FT SUPERSONIC	282
2422	PUBLISHED	FH15	5* 1*78 - 05*05*78	ET/SPIKE	(0)	AEDC A / SUPERSONIC	420
2423	PUBLISHED	FH16	7* 1*78 - 07*15*78	ET/SPIKE	(0)	ARC 3.5-FT HYPERSONIC	247
2424	PUBLISHED	OA126A	5* 1*78 - 05*30*78	5/140C	(47)	ARC 11-FT TRANSONIC	289
2425							
2426	PUBLISHED	LA124	6* 7*77 - 06*10*77	5/140C	(74)	LARC UNITARY PLAN	1207
2427	PUBLISHED	OH103B	4*27*78 - 04*28*78	5/140C	(60)	AEDC D / HYPERSONIC	V2C
2428	PUBLISHED	IH11	4* 1*78 - 04*18*78	5/140C	(84)	LERC 10X10-FT SUPERSONIC	045
2429	PUBLISHED	IH51B	7*15*78 - 07*24*78	FLAT PLATE	(58)	ARC 3.5-FT HYPERSONIC	239
2430	PUBLISHED	OA270A	5*15*78 - 06*09*78	VEH 102	(39)	LARC 16-FT TRANSONIC	325
2431	PUBLISHED	IH85	4*19*78 - 04*26*78	5/140C	(60)	AEDC A / SUPERSONIC	W5
2432	PUBLISHED	LA125	7* 3*78 - 07*05*78	VEH. 102	(105)	LARC UNITARY PLAN	1243
2433	PUBLISHED	OA171	6* 5*78 - 06*22*78	VEH. 102	(105)	NSWC HYPERSONIC LAB (#9)	1310
2434	PUBLISHED	OA129	7* 7*78 - 07*15*78	VEH102	(47)	AEDC 16-FT TRANSONIC	507
2435	PUBLISHED	IH39	9*22*76 - 04*14*77	5/140C	(19)	LERC 10X10-FT SUPERSONIC	041
2436							
2437	PUBLISHED	FA25	4*15*78 - 08*01*78	5/140C	(74)	MSFC 14-IN TRANSONIC	652
2438	PUBLISHED	IA138	8*21*78 - 09*01*78	5/140C	(75)	ARC 9X7-FT SUPERSONIC	246
2439	PUBLISHED	IA182	9*19*78 - 09*20*78	5/140C	(47)	AEDC 16-FT TRANSONIC	517
2440	PUBLISHED	IH83	1*25*78 - 03*10*78	5/140C	(19)	LERC 10X10-FT SUPERSONIC	044
2441							
2442							
2443	PUBLISHED	OH79	6* 1*78 - 08*24*78	5/140C	(65)	JSC VAC. CHAMBER A	61-A
2444	PUBLISHED	IA183	11*15*78 - 11*16*78	VEH 102	(89)	AEDC 16-FT TRANSONIC	519
2445	PUBLISHED	OA146	11*28*78 - 12*07*78	5/140C	(47)	ARC 8X7-FT SUPERSONIC	318

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2446								
2447								
2448	PUBLISHED	IH51C	12*26*78 -	02*16*79	FLAT PLATE	(58)	ARC 3.5-FT HYPERSONIC	241
2449	PUBLISHED	IA132	11*27*78 -	12*14*78	ET FORETANK	(68)	AEDC 16-FT TRANSONIC	505
2450	PUBLISHED	OS12	1*11*76 -	01*29*76	LRSI TILE	(85)	ARC 2X2-FT TRANSONIC	116
2451	PUBLISHED	OH90	3* 2*78 -	03*11*78	ELEV/ELEV	(94)	AEDC B / HYPERSONIC	P4A
2452	PUBLISHED	IH99	8*28*77 -	09*07*77	5/140C	(98)	ARC 3.5-FT HYPERSONIC	230
2453	PUBLISHED	IH75	10* 3*77 -	12*12*77	5/140C	(19)	CALSPAN 32-IN LUDWIEG	100
2454	PUBLISHED	LA57B	6* 4*75 -	06*06*75	140A,B	(0)	LARC 31-IN CONT-FLOW HYP.	114
2455	PUBLISHED	OH102A	10*25*78 -	11*29*78	5/140C	(56)	AEDC B / HYPERSONIC	B65
2456	PUBLISHED	IA184	4* 2*79 -	04*13*79	5/140C	(47)	ARC 9X7-FT SUPERSONIC	347
2457	PUBLISHED	IA180	3*26*79 -	03*30*79	ET FORETANK	(68)	LARC UNITARY PLAN	1267
2458	PUBLISHED	OS41	4*18*79 -	04*20*79	LRSI TILES	(96)	ARC 11-FT TRANSONIC	369
2459								
2460	IN PROCESS	FA27	3*14*79 -	05*16*79	5/140C	(74)	MSFC 14-IN TRANSONIC	655
2461	PUBLISHED	IH51D	5* 1*79 -	06*00*79	FLAT PLATE	(58)	ARC 3.5-FT HYPERSONIC	244
2462	PUBLISHED	IA131C	3* 5*79 -	03*11*79	ET FORETANK	(68)	ARC 8X7-FT SUPERSONIC	283
2463	PUBLISHED	OS42	7* 2*79 -	07*05*79	TPS TILES	(96)	ARC 11-FT TRANSONIC	380
2464	PUBLISHED	IH102-3	5* 1*79 -	06*01*79	5/140C	(83)	AEDC A / SUPERSONIC	B67
2465	PUBLISHED	OS57	8*26*81 -	08*27*81	TILE	(81)	ARC 9X7-FT SUPERSONIC	508
2466	PUBLISHED	OA257	3*12*81 -	04*20*81	VEH 102	(72)	LARC 20-IN HYPERSONIC (M=6)	6559
2467	PUBLISHED	IH103-2	10*15*79 -	11*01*79	5/140C	(56)	ARC 3.5-FT HYPERSONIC	245
2468	PUBLISHED	OH105B	7*23*79 -	08*01*79	5/140C	(60)	ARC 3.5-FT HYPERSONIC	247
2469								
2470	PUBLISHED	OS45	9* 3*79 -	09*03*79	TPS TILES	(96)	ARC 11-FT TRANSONIC	381

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2471	PUBLISHED	LA132	10*11*79	- 11*01*79	VEH 102	(89)	LARC 16-FT TRANSONIC	341
2472	PUBLISHED	OH400	8* 1*79	- 09*01*79	140C SILTS	(92)	ARC 11-FT TRANSONIC	B65
2473	PUBLISHED	OA252	10*16*79	- 11*14*79	TILE	(107)	ARC 2X2-FT TRANSONIC	382
2474	PUBLISHED	FA28	8* 1*79	- 09*01*79	5/140C	(74)	MSFC 14-IN TRANSONIC	656
2475	PUBLISHED	LA140	12*26*79	- 01*03*80	VEH. 102	(105)	LARC 16-FT TRANSONIC	342
2476	IN PROCESS	IA190B	5*20*80	- 02*21*80	5/140C	(47)	ARC 9X7-FT SUPERSONIC	411
2477	PUBLISHED	LA141B	3*18*80	- 05*01*80	VEH 102	(74)	LARC 20-IN HYPERSONIC (M=6)	6546
2478	PUBLISHED	LA131	1* 8*80	- 02*01*80	VEH 102	(106)	LARC UNITARY PLAN	1299
2479								
2480	PUBLISHED	IH104	2* 7*80	- 04*17*80	5/140C	(60)	ARC 3.5-FT HYPERSONIC	250
2481								
2482	PUBLISHED	OA400	4*23*80	- 05*02*80	5/140C	(47)	ARC 11-FT TRANSONIC	427
2483	PUBLISHED	OS49	1*28*81	- 02*04*81	TPS	(111)	AEDC 16-FT TRANSONIC	556
2484	PUBLISHED	LA144	7*28*80	- 08*01*80	VEH 102	(106)	LTV 4X4-FT SUPERSONIC	742
2485	PUBLISHED	OS50	4*30*81	- 06*01*81	TPS	(113)	ARC 11-FT TRANSONIC	425
2486	PUBLISHED	OA253	7* 1*80	- 07*08*80	5/140C	(84)	AEDC 16-FT TRANSONIC	574
2487	PUBLISHED	OS51C	1*27*81	- 01*29*81	TPS TILE	(96)	ARC 11-FT TRANSONIC	436
2488								
2489	PUBLISHED	OS56	8*26*81	- 08*27*81	TPS TILE	(108)	AEDC 16-FT TRANSONIC	608
2490	PUBLISHED	OH109	10*27*80	- 11*24*80	5/140C	(56)	AEDC B / HYPERSONIC	G9
2491	PUBLISHED	OA258	11*25*80	- 01*06*81	VEH 102	(106)	AEDC B / HYPERSONIC	BHQ
2492	PUBLISHED	OH107	1* 7*81	- 01*08*81	ELEV/ELEV	(94)	AEDC B / HYPERSONIC	B17
2493	PUBLISHED	OA259	2*16*81	- 02*20*81	VEH 102	(72)	AEDC B / HYPERSONIC	14
2494	PUBLISHED	OH108	12*15*80	- 01*15*81	ELEV/ELEV	(93)	ARC 3.5-FT HYPERSONIC	254
2495	PUBLISHED	OH110	11*17*80	- 01*30*81	5/140C	(60)	ARC 3.5-FT HYPERSONIC	253

DATAMAN NO.	DATAMAN STATUS	TEST NO.	TESTING SCHED.	TESTING COMPL.	MODEL REF.	MODEL (ID)	WIND TUNNEL FACILITY	WIND TUNNEL NO.
2495	PUBLISHED	OH111	9*24*81	09*30*81	5/140C	(60)	AEDC B / HYPERSONIC	1C
2497	IN PROCESS	MA34	3*12*81	03*20*81	ADS PROBES	(99)	AEDC 16-FT TRANSONIC	594
2498	PUBLISHED	OA255D	1*12*81	02*02*81	OV102	(70)	LARC UNITARY PLAN	1319
2499	PUBLISHED	OA164	11*28*75	12*01*75	OV101(ALT)	(76)	ARC 40X80-FT SUBSONIC	473
2500								
2501								
2502								
2503	PUBLISHED	OS53B	3*23*81	04*01*81	TPS	(719)	LARC 8-FT TRANSONIC PRESSURE	909
2504								
2505	PUBLISHED	OS46G	12*10*81	12*11*81	TPS	(108)	AEDC 16-FT TRANSONIC	551
2506	PUBLISHED	OS60	6* 9*81	06*09*81	TPS TILE	(96)	ARC 11-FT TRANSONIC	500
2507	PUBLISHED	MA33A	4*19*82	04*30*82	VEH 102	(106)	ARC 11-FT TRANSONIC	510
2508								
2509								
2510								

APPENDIX B

TABLE B2. - DATAMAN DOCUMENT TITLES

CHRYSLER DATA MANAGEMENT SERVICES(DMS)
SPACE SHUTTLE WIND TUNNEL TEST PROGRAM
DATA REPORT DOCUMENTATION

DMS-DR REPORT NUMBER	NASA CR NUMBER	NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2001	128,750	MA5	AERODYNAMIC STABILITY AND CONTROL CHARACTERISTICS OF A .01925 SCALE MODEL NR ATP ORBITER AT MACH NUMBERS FROM 1.9 TO 4.63
2002	128,752	LA1	RESULTS OF TRANSONIC TESTS IN THE NASA/LARC 8 FOOT PRESSURE TUNNEL ON A 0.015 SCALE MODEL NR-PRR SPACE SHUTTLE ORBITER
2003	128,754	MA2	HYPERSONIC AERODYNAMIC CHARACTERISTICS OF NR-ATP ORBITER, ORBITER WITH EXTERNAL TANK, AND ASCENT CONFIGURATION
2004	120,082	MA1	LONGITUDINAL AERODYNAMIC CHARACTERISTICS OF LOW ASPECT RATIO WING CONFIGURATIONS IN GROUND EFFECT FOR A MOVING AND STATIONARY GROUND SURFACE
2005	120,070	OA1	AERODYNAMIC STABILITY, CONTROL EFFECTIVENESS AND DRAG CHARACTERISTICS OF A SHUTTLE ORBITER CONFIGURATION AT MACH NUMBERS FROM 0.6 TO 4.96
2006	120,088	IA1A	AERODYNAMIC STATIC STABILITY AND CONTROL EFFECTIVENESS OF A PARAMETRIC SHUTTLE LAUNCH CONFIGURATION
2007	128,760	OA4	RESULTS OF INVESTIGATIONS ON A 0.015 SCALE MODEL NORTH AMERICAN ROCKWELL SPACE SHUTTLE ORBITER IN THE NASA/ARC 3.5 FOOT HYPERSONIC WIND TUNNEL
2008	128,751	MA4	STATIC STABILITY AND PERFORMANCE CHARACTERISTICS OF THE A.T.P. ORBITER AT M=10.3
2008 R-01	128,751	MA4	STATIC STABILITY AND PERFORMANCE CHARACTERISTICS OF THE A.T.P. ORBITER AT M=10.3
2009	128,761	OA3	AERODYNAMIC CHARACTERISTICS OF THE ROCKWELL INTERNATIONAL ORBITER OA3 AT MACH NUMBERS FROM 0.6 TO 2.0
2010	120,060	IA1B	DETERMINATION OF THE AERODYNAMIC INTERFERENCE BETWEEN THE SPACE SHUTTLE ORBITER, EXTERNAL TANK, AND SOLID ROCKET BOOSTER ON A 0.004 SCALE ASCENT CONFIGURATION
2011	120,089	MA9F	SPACE SHUTTLE (ATP CONFIGURATION) ABORT STAGING INVESTIGATION
2012	120,090	SA1F	AERODYNAMIC CHARACTERISTICS OF A 162-INCH DIAMETER SOLID ROCKET BOOSTER WITH AND WITHOUT STRAKES
2013	128,762	IA2	EFFECT OF GASEOUS AND SOLID SIMULATED JET PLUMES ON AN 040A SPACE SHUTTLE LAUNCH CONFIGURATION AT MACH NUMBERS FROM 1.6 TO 2.2

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2014	128,753		OA7	RESULTS OF SUPERSONIC TESTS IN THE LARC UNITARY PLAN WIND TUNNEL ON A .015 SCALE MODEL NR-PRR SPACE SHUTTLE ORBITER
2015	120,091	V-01	IA4	AERODYNAMIC RESULTS OF SEPARATION TESTS IN THE VOUGHT AERONAUTICS 4X4FT HSWT ON A .0075 SCALE ROCKWELL INTERNATIONAL-ATP SHUTTLE INTEGRATED VEHICLE
2015	120,091	V-02	IA4	AERODYNAMIC RESULTS OF SEPARATION TESTS ON THE VOUGHT AERONAUTICS 4FT X 4FT HSWT ON A .0075 SCALE ROCKWELL INTERNATIONAL-ATP SHUTTLE INTEGRATED VEHICLE
2016	120,092		OA2	RESULTS OF INVESTIGATIONS ON A 0.0405 SCALE MODEL ATP VERSION OF THENR-SSV ORBITER IN THE NORTH AMERICAN AERONAUTICAL LABORATORY LOW SPEED WIND TUNNEL
2017	123,851		OA5	RESULTS OF INVESTIGATIONS ON A 0.0405 SCALE MODEL PRR VERSION OF THENR-SSV ORBITER IN THE NORTH AMERICAN AERONAUTICAL LABORATORY LOW SPEED WIND TUNNEL
2018	128,755		IA3	CROSS WIND LOADS INVESTIGATION OF A .01925 SCALE MODEL OF THE ATP-SSV LAUNCH CONFIGURATION
2019	128,756		OA6	LOW SPEED LONGITUDINAL AND LATERAL STABILITY CHARACTERISTICS OF A PRPRR SHUTTLE ORBITER CONFIGURATION
2020	128,757		OA9	LOW SPEED INVESTIGATION OF THE PRR PLANFORM WING BOTH IN AND OUT OF GROUND EFFECT
2021	128,758	V-01	OA45	PRESSURE LOADS AND AERODYNAMIC FORCE INFORMATION FOR THE -89A SPACE SHUTTLE ORBITER CONFIGURATION
2021	128,758	V-02	OA45	PRESSURE LOADS AND AERODYNAMIC FORCE INFORMATION FOR THE -89A SPACE SHUTTLE ORBITER CONFIGURATION
2022	128,759		OA10	AERODYNAMIC CHARACTERISTICS OF THE ROCKWELL INTERNATIONAL -89B SPACE SHUTTLE ORBITER CONFIGURATION
2023	128,763		LA2	STATIC AERODYNAMIC CHARACTERISTICS AND OIL FLOW AND ELECTRON BEAM RESULTS OF A 0.005 SCALE MODEL LANGLEY CONCEPT SPACE SHUTTLE ORBITER(LO-100) AT A MACH NUMBER OF 20.3
2024	128,766		IA7	WIND TUNNEL TEST OF THE 0.019 (040A) JET PLUME SPACE SHUTTLE INTEGRATED VEHICLE IN THE ARC 11-FOOT UNITARY WIND TUNNEL
2025	128,767		SA3F	AERODYNAMIC CHARACTERISTICS OF A 142-INCH DIAMETER SOLID ROCKET BOOSTER WITH AND WITHOUT STRAKES

CHRYSLER DATA MANAGEMENT SERVICES(DMS)
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DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2026	128,778		IA31F	ARODYNAMIC INVESTIGATIONS ON A 0.004 SCALE MODEL MCR 0074 BASELINE SPACE SHUTTLE LAUNCH VEHICLE AT MACH NO. BETWEEN 0.6 AND 4.96
2027	141,807	V-01	IA32FB	AN INVESTIGATION IN THE NASA MSFC 14-INCH TRISONIC WIND TUNNEL TO DETERMINE THE PRESSURE DISTRIBUTION OVER THE COMPONENTS OF A 0.004 SCALE VERSION OF THE ROCKWELL MCR 0074 BASELINE SHUTTLE ASCENT CONFIGURATION (IA32FB)
2027	141,808	V-02	IA32FB	AN INVESTIGATION IN THE NASA MSFC 14-INCH TRISONIC WIND TUNNEL TO DETERMINE THE PRESSURE DISTRIBUTION OVER THE COMPONENTS OF A 0.004 SCALE VERSION OF THE ROCKWELL MCR 0074 BASELINE SHUTTLE ASCENT CONFIGURATION (IA32F)
2027	141,809	V-03	IA32FB	AN INVESTIGATION IN THE NASA MSFC 14-INCH TRISONIC WIND TUNNEL TO DETERMINE THE PRESSURE DISTRIBUTION OVER THE COMPONENTS OF A 0.004 SCALE VERSION OF THE ROCKWELL MCR 0074 BASELINE SHUTTLE ASCENT CONFIGURATION (IA32F)
2028	134,434	V-01	IA31FB	TRIPLE BALANCE TEST OF THE PRR BASELINE SPACE SHUTTLE CONFIGURATION (TWT 570)
2028	134,436	V-02	IA31FB	TRIPLE BALANCE TEST OF THE PRR BASELINE SPACE SHUTTLE CONFIGURATION (TWT 570)
2029	128,765		OA47	RESULTS OF A STATIC STABILITY AND CONTROL EFFECTIVENESS INVESTIGATION OF A 0.004 SCALE 2A ORBITER IN THE MARSHALL SPACE FLIGHT CENTER TRISONIC WIND TUNNEL (MACH=0.6-4.96)
2030	128,768		OA14	AERODYNAMIC CHARACTERISTICS OF VARIOUS AFT-END CONFIGURATIONS OF THE ROCKWELL INTERNATIONAL -89B SPACE SHUTTLE ORBITER
2031	128,769		LA3	HYPERSONIC PERFORMANCE, STABILITY AND CONTROL CHARACTERISTICS OF A 0.010 SCALE MODEL OF A LANGLEY CONCEPT SPACE SHUTTLE ORBITER
2032	128,794	V-01	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-02	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-03	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS

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DATA REPORT DOCUMENTATION

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2032	128,794	V-04	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-05	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-06	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-07	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-08	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-09	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-10	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-11	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-12	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-13	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2032	128,794	V-14	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-15	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-16	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-17	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2032	128,794	V-18	IA9A,B,C/OA12A,C	RESULTS OF TESTS OA12 AND IA9 IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS ON AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 2A TO DETERMINE AERODYNAMIC LOADS
2033	128,772		LA4	SUPERSONIC STABILITY AND CONTROL CHARACTERISTICS OF A LANGLEY CONCEPT SPACE SHUTTLE ORBITER AT MACH 1.5 TO 4.63
2034	128,764		LA22	AERODYNAMIC AND FLOW VISUALIZATION STUDIES ON A SPACE SHUTTLE CONCEPT WITH A DOUBLE DELTA WING ORBITER AT A MACH NUMBER OF 20.3
2035	134,077		OH2A/OH2B	THERMAL PROTECTION SYSTEM GAP HEATING RATES OF THE ROCKWELL INTERNATIONAL FLAT PLATE HEAT TRANSFER MODEL
2036	128,775		LA5	AERODYNAMIC AND FLOW-VISUALIZATION STUDIES ASSOCIATED WITH VARIATIONS IN THE GEOMETRY OF THE FORWARD PORTION OF IRREGULAR PLANFORM WINGSAT A MACH NUMBER OF 20.3
2037	134,405		OA84	RESULTS OF INVESTIGATIONS ON A 0.015-SCALE 140A/B CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL (49-O) IN THE LTV 4 BY 4-FOOT HIGH SPEED WIND TUNNEL
2038	128,793		OA16	RESULTS OF LOW SPEED WIND TUNNEL TESTS ON A .0405 SCALE MODEL ROCKWELL SPACE SHUTTLE ORBITER TESTED BOTH IN FREE AIR AND IN THE PRESENCE OF A GROUND PLANE
2039	134,071		IA6A	RESULTS OF WIND TUNNEL TESTS AT MACH 5 ON THE .004 SCALE MODEL 2A CONFIGURATION SPACE SHUTTLE TO DETERMINE PROXIMITY EFFECTS AND ORBITERCONTROL EFFECTIVENESS DURING ORBITER/EXTERNAL TANK ABORT SEPARATION

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DATA REPORT DOCUMENTATION

DMS-DR REPORT NUMBER	NASA CR NUMBER	NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2040	128,773	LA6	SURFACE ROUGHNESS EFFECTS ON THE TRANSONIC AERODYNAMICS OF THE ROCKWELL INTERNATIONAL 089B-139 ORBITER
2041	128,781	LA7A	TRANSONIC AERODYNAMIC CHARACTERISTICS ASSOCIATED WITH VARIATIONS IN THE GEOMETRY OF THE FORWARD PORTION OF IRREGULAR PLANFORM WINGS
2042	134,087	IA52	RESULTS OF FLOW VISUALIZATION STUDIES IN THE NASA/MSFC 14 X 14 INCH TRISONIC WIND TUNNEL ON A .004 SCALE MODEL (34-O) SPACF SHUTTLE ORBITER AND INTEGRATED VEHICLE
2043	128,770	LA16	HEAT TRANSFER DATA TO CAVITIES BETWEEN SIMULATED RSI TILES AT MACH 8
2044	128,786	OA11A	RESULTS OF INVESTIGATIONS ON A 0.015-SCALE MODEL 2A CONFIGURATION OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER IN THE NASA/AMES RESEARCH CENTER 3.5 FOOT HYPERSONIC WIND TUNNEL
2045	128,779	OA18	RESULTS OF INVESTIGATIONS (OA18) OF A 0.0405 SCALE MODEL OF THE 2A AND 3 SPACE SHUTTLE ORBITER CONFIGURATIONS IN THE NORTH AMERICAN AERONAUTICAL LABORATORY LOW SPEED WIND TUNNEL AT M = 0.26 AND 0.16
2046	128,776	LA17	AERODYNAMIC STABILITY AND CONTROL CHARACTERISTICS OF A LANGLEY CONCEPT SPACE SHUTTLE ORBITER (LO-100) AT MACH NUMBERS OF 0.35 TO 1.2
2047	134,086	LA31	EFFECT OF WALL TO TOTAL TEMPERATURE RATIO VARIATION ON HEAT TRANSFER
2048	134,104	IA12B	WIND TUNNEL TEST OF THE 0.019 (2A CONFIGURATION) JET PLUME SPACE SHUTTLE INTEGRATED VEHICLE IN THE ARC 9- BY 7-FOOT UNITARY WIND TUNNEL
2049	128,771	OH40	AERODYNAMIC HEATING OF A SPACE SHUTTLE DOUBLE DELTA WING ORBITER AT MACH NUMBER 8.0
2050	128,790	OA43	WIND TUNNEL TEST OF THE 0.15-SCALE ROCKWELL INTERNATIONAL SPACE SHUTTLE VEHICLE ORBITER IN THE AMES 6- BY 6-FOOT SUPERSONIC WIND TUNNEL
2051	128,774	SA5F	AERODYNAMIC CHARACTERISTICS OF A 142-INCH DIAMETER SOLID ROCKET BOOSTER (CONFIGURATIONS 89B AND 139)
2052	128,791	LA10	SUPERSONIC AERODYNAMIC CHARACTERISTICS ASSOCIATED WITH VARIATIONS INTHE GEOMETRY OF THE FORWARD PORTION OF IRREGULAR PLANFORM WINGS
2053	128,792 V-01	OA21B	EXPERIMENTAL INVESTIGATIONS OF AN 0.0405 SCALE SPACE SHUTTLE CONFIGURATION 3 ORBITER TO DETERMINE SUBSONIC STABILITY CHARACTERISTICS (OA21)

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2053	128,792	V-02	OA21B	EXPERIMENTAL INVESTIGATIONS OF AN 0.0405 SCALE SPACE SHUTTLE CONFIGURATION 3 ORBITER TO DETERMINE SUBSONIC STABILITY CHARACTERISTICS (OA21)
2054	128,796		LA8A/LA8B	SURFACE ROUGHNESS EFFECTSON THE SUPERSONIC AERODYNAMICS OF THE ROCKWWELL INTERNATIONAL 089B-139 ORBITER
2055	128,780	V-01	OA48	STATIC STABILITY AND CONTROL EFFECTIVENESS OF MODELS 12-0 AND 34-0 OF THE VEHICLE 3 CONFIGURATIONS
2055	128,780	V-02	OA43	STATIC STABILITY AND CONTROL EFFECTIVENESS OF MODELS 12-0 AND 34-0 OF THE VEHICLE 3 CONFIGURATIONS
2055	128,780	V-03	OA48	STATIC STABILITY AND CONTROL EFFECTIVENESS OF MODELS 12-0 AND 34-0 OF THE VEHICLE 3 CONFIGURATIONS
2056	128,782		LA9	SURFACE ROUGHNESS EFFECTS ON THE SUBSONIC AERODYNAMICS OF THE ROCKWELL INTERNATIONAL 089B-139 ORBITER
2057	134,411		OA44	RESULTS OF AN EXPERIMENTAL AERODYNAMIC INVESTIGATION TO OBTAIN STATIC STABILITY AND CONTROL CHARACTERISTICS OF THE SSV CONFIGURATIONS 2A(VL70-000089B) MODEL 1 AND 3 (VL70-000139B) MODEL 2 ORBITERS AT MACHNUMBERS OF 2.5, 3.9, AND 4.6 IN THE NASA LARC 4X4-FOOT UPWT (OA44)
2058	134,079		OA17	RESULTS OF THE 0.015 SCALE SPACE SHUTTLE VEHICLE ORBITER TEST (OA17)IN THE NASA LOW TURBULENCE PRESSURE TUNNEL
2059	128,798		OA11B	INVESTIGATIONS OF THE SPACE SHUTTLE ORBITER 2A CONFIGURATION 0.015-SCALE MODEL IN THE NASA AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL AT MACH NUMBERS 5, 7 AND 10
2060	134,091		OA58	RESULTS OF AN AERODYNAMIC FORCE AND MOMENT INVESTIGATION OF AN 0.015-SCALE CONFIGURATION 3 SPACE SHUTTLE ORBITER IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (OA58)
2061	128,789		OA68	SUBSONIC, TRANSONIC, AND SUPERSONIC STABILITY AND CONTROL CHARACTER-ISTICS OF THE -147B SPACE SHUTTLE ORBITER
2062	134,117	V-01	IA13	AERODYNAMIC RESULTS OF A SEPARATION EFFECTS TEST CONDUCTED IN THE AEDC 40X 40 INCH TUNNEL A FACILITY ON THE ROCKWELL INTERNATIONAL LAUNCH CONFIGURATION 3 INTEGRATED VEHICLE

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2062	134,118	V-02	IA13	AERODYNAMIC RESULTS OF A SEPARATION EFFECTS TEST CONDUCTED IN THE AEDC 40 X 40 INCH TUNNEL A FACILITY ON THE ROCKWELL INTERNATIONAL LAUNCH CONFIGURATION 3 INTEGRATED VEHICLE
2062	141,801	V-03	IA13	AERODYNAMIC RESULTS OF A SEPARATION EFFECTS TEST CONDUCTED IN THE AEDC 40 X 40 INCH TUNNEL A FACILITY ON THE ROCKWELL INTERNATIONAL LAUNCH CONFIGURATION 3 INTEGRATED VEHICLE
2063	128,788		IA37/IA48	RESULTS OF TESTS IN THE MSFC 14X14 INCH TRISONIC WIND TUNNEL ON A .004 SCALE MODEL OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE VEHICLE 3. (INTEGRATED CONFIGURATION)
2064	141,814	V-01	IA36	WIND TUNNEL TEST OF THE 0.019 SCALE SPACE SHUTTLE INTEGRATED VEHICLE(MODEL 14-OTS) IN THE CALSPAN 8-FOOT TRANSONIC WIND TUNNEL (IA36)
2064	141,816	V-02	IA36	WING TUNNEL TEST OF THE 0.019 SCALE SPACE SHUTTLE INTEGRATED VEHICLE(MODEL 14-OTS) IN THE CALSPAN 8-FOOT TRANSONIC WIND TUNNEL (IA36)
2065	141,518	V-01	IA12C	WIND TUNNEL TESTS OF AN 0.019-SCALE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA AMES 8 X 7-FOOT UNITARY WIND TUNNEL(IA12C)
2065	141,519	V-02	IA120	WIND TUNNEL TESTS OF AN 0.019-SCALE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA AMES 8 X 7-FOOT UNITARY WIND TUNNEL(IA12C)
2065	141,520	V-03	IA12C	WIND TUNNEL TESTS OF AN 0.019-SCALE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA AMES 8 X 7-FOOT UNITARY WIND TUNNEL(IA12C)
2066	128,783		LA11	HYPERSONIC PERFORMANCE, STABILITY AND CONTROL CHARACTERISTICS OF A .0075 SCALE MODEL ROCKWELL INTERNATIONAL 089-139 ORBITER CONFIGURATION
2067	128,777		OS2	FLUTTER TESTS (OS2) OF THE SHUTTLE ORBITER FIN/RUDDER MODEL 24-O
2068	128,797		OA71A	EFFECTS OF THE AIR BREATHING PROPULSION SYSTEM ON SPACE SHUTTLE ORBITER SUBSONIC STABILITY AND CONTROL CHARACTERISTICS (OA71A)
2069	134,074		MA7	EFFECTS OF REACTION CONTROL SYSTEM JET-FLOW FIELD INTERACTIONS ON A 0.015 SCALE MODEL SPACE SHUTTLE ORBITER AERODYNAMIC CHARACTERISTICS
2070	128,787		LA23	EFFECT OF GASEOUS AND SOLID SIMULATED JET PLUMES ON AN O40A SPACE SHUTTLE LAUNCH CONFIGURATION AT MACH NUMBERS FROM 1.6 TO 2.2

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2071	128,799		OA23	RESULTS OF TESTS OF 0.010- AND 0.015-SCALE MODELS OF SPACE SHUTTLE ORBITER CONFIGURATIONS 3 AND 3A IN THE AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL (OA23)
2072	134,072		IA31FC	MISALIGNMENT STUDIES ON SPACE SHUTTLE INTEGRATED VEHICLE
2073	134,070		OA70	EFFECTS OF REACTION CONTROL SYSTEM JET SIMULATION ON THE STABILITY AND CONTROL CHARACTERISTICS OF A 0.015 SCALE SPACE SHUTTLE MODEL TESTED IN THE LANGLEY RESEARCH CENTER UNITARY PLAN WIND TUNNEL
2074	134,414		OA57A	EFFECTS OF THE AIR BREATHING ENGINE PLUMES ON SSV ORBITER SUBSONIC WING PRESSURE DISTRIBUTIONS
2075	128,784		OH41	INVESTIGATION OF CONFIGURATION EFFECTS ON ENTRY HEATING DISTRIBUTIONS AT MACH = 8.0 (OH41)
2076	128,785		OH41A	INVESTIGATION OF CONFIGURATION EFFECTS ON ENTRY HEATING DISTRIBUTIONS AT MACH NO = 8.0 (OH41A)
2077	134,095	V-01	IA29/OA63	RESULTS OF TESTS OA63 AND IA29 ON AN 0.015-SCALE MODEL OF THE SPACE SHUTTLE CONFIGURATION 140 A/B IN THE NASA/ARC 6- BY 6-FOOT TRANSONIC WIND TUNNEL
2077	134,099	V-02	IA29	RESULTS OF TESTS OA63 AND IA29 ON AN 0.015-SCALE MODEL OF THE SPACE SHUTTLE CONFIGURATION 140 A/B IN THE NASA/ARC 6- BY 6-FOOT TRANSONIC WIND TUNNEL
2077	134,100	V-03	OA63	RESULTS OF TESTS OA63 AND IA29 ON AN 0.015-SCALE MODEL OF THE SPACE SHUTTLE CONFIGURATION 140 A/B IN THE NASA/ARC 6- BY 6-FOOT TRANSONIC WIND TUNNEL
2078	128,795		IA10	WIND TUNNEL TEST OF THE 0.010-SCALE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA-AMES 3.5-FOOT HYPERSONIC WIND TUNNEL (IA10)
2079	134,083		LA15	EFFECTS OF SURFACE ROUGHNESS ON THE AERODYNAMIC CHARACTERISTICS OF THE MODIFIED 089 B SHUTTLE ORBITER AT MACH 6 (LA15)
2080	134,416	V-01	OA57B	EFFECTS OF AIR BREATHING ENGINE PLUMES ON SSV ORBITER SUBSONIC WING PRESSURE DISTRIBUTION
2080	134,417	V-02	OA57B	EFFECTS OF AIR BREATHING ENGINE PLUMES ON SSV ORBITER SUBSONIC WING PRESSURE DISTRIBUTION
2081	141,580	V-01	OA69	LANDING PRESSURE LOADS OF THE -140 A/B SPACE SHUTTLE ORBITER DETER- MINED IN THE NRLAD LOW SPEED WIND TUNNEL (OA69)

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2081	141,581	V-02	OA69	LANDING PRESSURE LOADS OF THE -140 A/B SPACE SHUTTLE ORBITER DETER- MINED IN THE NRLAD LOW SPEED WIND TUNNEL (OA69)
2082	128,800		OA73	EFFECTS OF REACTION CONTROL SYSTEM JET SIMULATION ON THE STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE SPACE SHUTTLE ORBITER MODEL IN THE AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL
2083	134,081		OA20A	RESULTS OF INVESTIGATIONS (OA20) ON A 0.015-SCALE 140 A/B CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL IN THE NASA/LANGLEY RESEARCH CENTER UNITARY PLAN WIND TUNNEL
2084	134,443	V-01	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLEVEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA'4A)
2084	134,444	V-02	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLEVEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)
2084	143,445	V-03	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)
2084	143,446	V-04	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)
2084	143,447	V-05	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)
2084	143,448	V-06	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)
2084	143,449	V-07	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)
2084	143,450	V-08	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)

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2084	141,501	V-09	IA14A	IARLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)
2084	141,502	V-10	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)
2084	141,503	V-11	IA14A	AIRLOADS INVESTIGATIONS OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH RANGE 0.6 TO 1.4 (IA14A)
2085	167,344		OH10/IH2	REPORT OF PRESSURE DISTRIBUTION TESTS OF THE 0.010-SCALE SPACE SHUTTLE VEHICLE MODEL (26-OTS) IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (TESTS OH10 AND IH2)
2086	134,078		OA71C	EFFECTS OF THE SIX ENGINE AIR BREATHING PROPULSION SYSTEM ON SPACE SHUTTLE ORBITER SUBSONIC STABILITY AND CONTROL CHARACTERISTICS
2087	134,116		SA10F	EFFECT OF ENGINE SHROUD CONFIGURATION ON THE STATIC AERODYNAMIC CHARACTERISTICS OF A 0.00563 SCALE 142-INCH DIAMETER SOLID ROCKET BOOSTER
2088	134,105		SA2FA/SA2FB	AERODYNAMIC CHARACTERISTICS OF A 142-INCH DIAMETER SOLID ROCKET BOOSTER (CONFIGURATION 139)
2089	134,082		OA25	RESULTS OF INVESTIGATIONS ON AN 0.015-SCALE CONFIGURATION 140A/B SPACE SHUTTLE ORBITER MODEL (49-O) IN THE NASA/LANGLEY RESEARCH CENTER 8-FOOT TRANSONIC PRESSURE TUNNEL (OA25)
2090	134,080		LA8C	SUPERSONIC PERFORMANCE, STABILITY AND CONTROL CHARACTERISTICS OF A 0.01875 SCALE MODEL ROCKWELL INTERNATIONAL 089B-139B ORBITER CONFIGURATION (LA8C)
2091	141,512		LA7B	SUBSONIC AND TRANSONIC AERODYNAMIC CHARACTERISTICS ASSOCIATED WITH VARIATIONS IN THE GEOMETRY OF THE FORWARD PORTION OF IRREGULAR PLANFORM WINGS ON A .01875 SCALE LO-100 LANGLEY CONCEPT SPACE SHUTTLE ORBITER IN THE LANGLEY 8-FOOT TPT (LA7B)
2092	TM-X71968		OA72	HYPERSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.004 SCALE MODEL (34-O) ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER VEHICLE 3 CONFIGURATION (OA-72)
2093	134,090		IA37B	EFFECT OF EXTERNAL TANK NOSE SHAPE ON THE ROCKWELL INTERNATIONAL SPACE SHUTTLE VEHICLE 3. (INTEGRATED CONFIGURATION (IA37B)

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2094	134,073		OS1	FLUTTER TESTS (OS1) OF THE 0.02-SCALE ORBITER WING ELEVON SEMI-SPAN MODEL 23-O
2095	134,404		OA49	AN INVESTIGATION OF THE STABILITY AND CONTROL CHARACTERISTICS OF THE VEHICLE 4 CONFIGURATION
2096	134,101		OH13	HEAT TRANSFER TESTS OF AN 0.006-SCALE THIN SKIN SPACE SHUTTLE THERMOCOUPLE MODEL (41-O) IN THE LANGLEY RESEARCH CENTER VARIABLE DENSITY TUNNEL AT M=8
2097	134,102		OA62A	CONTINUED INVESTIGATIONS IN THE NAAL LOW SPEED WIND TUNNEL INTO THE EFFECTS OF THE AIR BREATHING PROPULSION SYSTEM ON ORBITER SUBSONIC STABILITY AND CONTROL CHARACTERISTICS (OA62A)
2098	134,096		'H15	HEAT TRANSFER TESTS OF A 0.006-SCALE THIN-SKIN SPACE SHUTTLE MODEL (41-OTS) IN THE AMES 3.5-FOOT HWT AT M=5.3
2099	134,419	V-01	OH4B	DATA REPORT FOR TESTS ON THE HEAT TRANSFER EFFECTS OF THE 0.0175-SCALE ROCKWELL INTERNATIONAL SPACE SHUTTLE VEHICLE MODEL 22-OT IN THE AEDC 50-INCH B WIND TUNNEL
2099	134,438	V-02	OH4B	DATA REPORT FOR TESTS ON THE HEAT TRANSFER EFFECTS OF THE 0.0175-SCALE ROCKWELL INTERNATIONAL SPACE SHUTTLE VEHICLE MODEL 22-OT IN THE AEDC 50-INCH WIND TUNNEL
2099	134,439	V-03	OH4B	DATA REPORT FOR TESTS ON THE HEAT TRANSFER EFFECTS OF THE 0.0175-SCALE ROCKWELL INTERNATIONAL SPACE SHUTTLE VEHICLE MODEL 22-OT IN THE AEDC 50-INCH B WIND TUNNEL
2100	134,075		OH3A/OH3B	PHASE CHANGE PAINT TESTS ON ROCKWELL ORBITER/TANK AND ORBITER ALONE CONFIGURATIONS
2101	134,076		OH42A/OH42B/OH42C	HEAT TRANSFER PHASE CHANGE PAINT TEST (OH-42) OF A ROCKWELL INTERNATIONAL SSV ORBITER IN THE NASA/LRC MACH 8 VARIABLE DENSITY WIND TUNNEL
2102	134,089		IA15	RESULTS OF INVESTIGATIONS ON A 0.010-SCALE MODEL OF THE CONFIGURATION 3 SPACE SHUTTLE ORBITER AND EXTERNAL TANK IN THE NASA/AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL (IA15)
2103	134,094		IA62F	WIND TUNNEL TEST RESULTS OF FAIRINGS ON A 0.004 SCALE MODEL ROCKWELLSPACE SHUTTLE INTEGRATED VEHICLE AERODYNAMIC CHARACTERISTICS AT MACHNUMBERS FROM 0.6 TO 4.96 (IA62F)

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2104	134,112	V-01	OA62B	INVESTIGATION OF SPACE SHUTTLE ORBITER SUBSONIC STABILITY AND CONTROL CHARACTERISTICS IN THE NAAL LOW SPEED WIND TUNNEL (OA62B)
2104	134,113	V-02	OA62B	INVESTIGATION OF SPACE SHUTTLE ORBITER SUBSONIC STABILITY AND CONTROL CHARACTERISTICS IN THE NAAL LOW SPEED WIND TUNNEL (OA62B)
2105	144,594		IH17	TRANSITION HEATING RATES OBTAINED ON A MATED AND ISOLATED 0.006 SCALE MODEL (41-OT) SPACE SHUTTLE ORBITER AND EXTERNAL TANK IN THE NASA/LARC VARIABLE DENSITY HYPERSONIC TUNNEL
2106	TM-X72630		LA14A/LA14B	SUPERSONIC DYNAMIC STABILITY DERIVATIVES OF A MODIFIED 089B SHUTTLE ORBITER
2107	TM-X72631		LA20	SUBSONIC AND TRANSONIC DYNAMIC STABILITY DERIVATIVES OF A MODIFIED 089B SHUTTLE ORBITER
2108	134,084		IA35/OA64	RESULTS OF TESTS (OA64 AND IA35) OF AN 0.015-SCALE MODEL (36-OTS) OF THE SPACE SHUTTLE CONFIGURATION 140A/B IN THE NASA/LARC UNITARY PLAN WIND TUNNEL
2109	141,527		OH45	ENTRY HEAT TRANSFER TESTS OF THE 0.006-SCALE SPACE SHUTTLE (-147B) ORBITER MODEL (50-O) IN THE LANGLEY RESEARCH CENTER FREON TUNNEL AT MACH 6 (OH45)
2110	144,589		IH18	HEAT TRANSFER TESTS OF AN 0.006-SCALE THIN-SKIN SPACE SHUTTLE THERMOCOUPLE MODEL (41-OT) IN THE LANGLEY RESEARCH CENTER FREON TUNNEL AT M = 6 (IH18)
2111	134,435		SA26F	REENTRY AERODYNAMIC CHARACTERISTICS OF A SPACE SHUTTLE SOLID ROCKET BOOSTER MODEL 449 TESTED IN MSFC 14 X 14 INCH TWT
2112	134,401		IA57	AERODYNAMIC RESULTS OF WIND TUNNEL SEPARATION TESTS ON A 0.01-SCALE MODEL (32-OTS) SPACE SHUTTLE INTEGRATED VEHICLE (IA57)
2113	134,111		OA85	EFFECTS OF REACTION CONTROL SYSTEM JET FLOW FIELD INTERACTIONS ON THE AERODYNAMIC CHARACTERISTICS OF A 0.010 SCALE SPACE SHUTTLE ORBITER MODEL IN THE LANGLEY RESEARCH CENTER 31-INCH CFHT
2114	134,098		OA86	AERODYNAMIC INVESTIGATIONS INTO VARIOUS LOW SPEED L/D IMPROVEMENT DEVICES ON THE 140A/B SPACE SHUTTLE ORBITER CONFIGURATION IN THE RI NAAL WIND TUNNEL (OA86)
2115	134,085		OA87	RESULTS OF INVESTIGATIONS ON A 0.015-SCALE MODEL (49-O) OF THE SPACE SHUTTLE ORBITER IN THE NASA/AMES 3.5-FOOT HYPERSONIC WIND TUNNEL (OA87)

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2116	134,888	OA91	EFFECT OF THE SIX ENGINE AIR BREATHING PROPULSION SYSTEM ON SPACE SHUTTLE ORBITER SUBSONIC AND TRANSONIC STABILITY AND CONTROL CHARACTERISTICS (OA91)
2117	147,617	OH14	TRANSITION HEATING RATES DETERMINED ON A 0.006 SCALE SPACE SHUTTLE ORBITER MODEL (NO. 50-0) IN THE NASA/LARC MACH 8 VARIABLE DENSITY WIND TUNNEL TEST (OH14)
2118	134,108	IA41	RESULTS OF TRANSONIC WIND TUNNEL TESTS ON AN 0.015 SCALE SPACE SHUTTLE MATED VEHICLE MODEL(67-OTS) IN THE LARC 8-FOOT TPT (IA41)
2119	134,109	IA42A/IA42B	SUPERSONIC TESTS OF AN 0.015-SCALE SPACE SHUTTLE MATED VEHICLE MODEL (67-OTS) IN THE LARC UPWT TO OBTAIN AERODYNAMIC FORCE DATA
2120	134,426	OA106	WIND TUNNEL TESTS OF AN 0.015-SCALE CONFIGURATION 140A/B SPACE SHUTTLE ORBITER MODEL (67-O) IN THE NASA/LRC 8-FOOT TPT TO OBTAIN TRANSONIC AERODYNAMIC FORCE DATA (OA106)
2121	TASK CANCELLED	LA38A	TRANSONIC AERODYNAMIC INVESTIGATION OF CONFIGURATION MODIFICATIONS TO RI-140A/B FOR EXTENDING CENTER OF GRAVITY RANGE
2122	134,424	IA69	INVESTIGATION OF SPACE SHUTTLE LAUNCH VEHICLE EXTERNAL TANK NOSE CONFIGURATION EFFECTS (MODEL 67-OTS) IN THE ROCKWELL INTERNATIONAL 7- BY 7-FOOT TRISONIC WIND TUNNEL (IA69)
2123	141,504	IA53	RESULTS FROM INVESTIGATIONS IN THE NASA/MSFC TWT ON A 0.004 SCALE MODEL SPACE SHUTTLE LAUNCH VEHICLE (MODEL 13P-OTS) TO DETERMINE GAS SUPPLY STRUT EFFECT ON MODEL PRESSURE ENVIRONMENT (IA53)
2124	134,093	IA16/OA26	RESULTS OF TESTS OA26 AND IA16 IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL ON A 0.015 SCALE MODEL (36-OTS) OF THE SPACE CONFIGURATION 140A/B TO OBTAIN PRESSURES FOR VENTING ANALYSIS
2125	134,409	OA88	HYPERSONIC STABILITY AND CONTROL CHARACTERISTICS AND REYNOLDS NUMBEREFFECTS OF THE ROCKWELL SSV 140 A/B ORBITER CONFIGURATION
2126	TASK CANCELLED	LA25	EFFECTS OF REACTION CONTROL SYSTEM JET SIMULATION ON THE HYPERSONIC PERFORMANCE, STABILITY AND CONTROL CHARACTERISTICS OF A .01 SCALE ROCKWELL INTERNATIONAL 139B ORBITER CONFIGURATION
2127	TM-X71954	LA35	REYNOLDS NUMBER EFFECTS AT MACH NUMBER 10.3 ON AERODYNAMIC CHARACTERISTICS OF .01 SCALE 139-B ORBITER

CHRYSLER DATA MANAGFMENT SERVICES(DMS)
SPACE SHUTTLE WIND TUNNEL TEST PROGRAM
DATA REPORT DOCUMENTATION

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2128	134,114	V-01	OA53A	INVESTIGATIONS ON AN 0.030-SCALE SPACE SHUTTLE VEHICLE CONFIGURATION140A/B ORBITER MODEL IN THE AMES RESEARCH CENTER11-BY11-FOOT SUPER- SONIC WIND TUNNEL (OA53A)
2128	134,115	V 02	OA53A	INVESTIGATIONS ON AN 0.030-SCALE SPACE SHUTTLE VEHICLE CONFIGURATION140A/B ORBITER MODEL IN THE AMES RESEARCH CENTER 11-BY11-FOOT SUPER-SONIC WIND TUNNEL (OA53A)
2129	141,522	V-01	IA14B	AIRLOADS INVESTIGATION OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 9- BY 7-FOOT UNITARY PLAN WIND TUNNEL FOR MACH 1.55 AND 2.2 (IA14B)
2129	141,523	V-02	IA14B	AIRLOADS INVESTIGATION OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B LAUNCH CONFIGURATION (MODEL 47-OTS) IN THE ARC 9- BY 7-FOOT UNITARY PLAN WIND TUNNEL FOR MACH 1.55 AND 2.2 (IA14B)
2130	141,529		OA22A	AIRLOADS INVESTIGATION OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 140A/B ORBITER CONFIGURATION (MODEL 47-O) IN THE ARC 11-FOOT UNITARY PLAN WIND TUNNEL FOR MACH 0.6 AND 0.9 (OA22A)
2131	141,530		OA22B	AIRLOADS INVESTIGATION OF AN 0.030-SCALE MODEL OF THE SPACE SHUTTLE 140A/B ORBITER CONFIGURATION (MODEL 47-O) IN THE ARC 9- BY 7-FOOT UNITARY PLAN WIND TUNNEL FOR MACH 1.55 AND 2.2 (OA22B)
2132	141,535		LA42	RESULTS OF DYNAMIC STABILITY TESTS CONDUCTED ON A .012 SCALE MODIFIED 089 B SHUTTLE ORBITER IN THE AEDC-VKF TUNNEL B AT A MACH NUMBER OF 8.0 (LA42)
2133	134,110		IA58	RESULTS OF TESTS IN THE NASA/LARC 31-INCH CFHT ON AN 0.010-SCALE MODEL (32-OT) OF THE SPACE SHUTTLE CONFIGURATION 3 TO OBTAIN HYPERSONIC AERODYNAMIC CHARACTERISTICS FOR SECOND STAGE OPERATION DURING NOMINAL BOOST AND THE ABORT RTLS MODE
2134 R-01	134,429		OA77/OA78	RESULTS OF INVESTIGATIONS (OA77 AND OA78) ON AN 0.015-SCALE 140A/B CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 49-0 IN THE AEDC VKF B AND C WIND TUNNELS
2135	TASK CANCELLED		LA13	
2136	141,514	V-01	IH3	RESULTS OF HEAT TRANSFER TESTS OF AN 0.0175-SCALE SPACE SHUTTLE VEHICLE MODEL 22 OTS IN THE NASA-AMES 3.5-FOOT HYPERSONIC WIND TUNNEL (IH3)

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2136	141,515	V-02	IH3	RESULTS OF HEAT TRANSFER TESTS OF AN 0.0175-SCALE SPACE SHUTTLE VEHICLE MODEL 22 OTS IN THE NASA-AMES 3.5-FOOT HYPERSONIC WIND TUNNEL (IH3)
2136	141,516	V-03	IH3	RESULTS OF HEAT TRANSFER TESTS OF AN 0.0175-SCALE SPACE SHUTTLE VEHICLE MODEL 22 OTS IN THE NASA-AMES 3.5-FOOT HYPERSONIC WIND TUNNEL (IH3)
2136	141,517	V-04	IH3	RESULTS OF HEAT TRANSFER TESTS OF AN 0.0175-SCALE SPACE SHUTTLE VEHICLE MODEL 22 OTS IN THE NASA-AMES 3.5-FOOT HYPERSONIC WIND TUNNEL (IH3)
2137	134,106	V-02	OA105	RESULTS OF TESTS IN THE NASA/LARC 31-INCH CFHT ON A 0.01-SCALE MODEL (32-OT) OF THE SPACE SHUTTLE CONFIGURATION 3 TO DETERMINE THE RCS JET FLOWFIELD INTERACTION EFFECTS ON AERODYNAMIC CHARACTERISTICS(IA60/OA105)
2137 R-01	134,103	V-01	IA60	RESULTS OF TESTS IN THE NASA/LARC 31-INCH CFHT ON A 0.01-SCALE MODEL (32-OT) OF THE SPACE SHUTTLE CONFIGURATION 3 TO DETERMINE THE RCS JET FLOWFIELD INTERACTION EFFECTS ON AERODYNAMIC CHARACTERISTICS(IA60/OA105) VOLUME 1 OF 2
2138	144,608	V-01	IH4	AEROHEATING(PRESSURE) CHARACTERISTICS OF A 0.010-SCALE VERSION OF THE VEHICLE 3 SPACE SHUTTLE CONFIGURATION(26-OTS) IN THE LANGLEY RESEARCH CENTER 4-FOOT WIND TUNNEL(IH4)
2138	144,609	V-02	IH4	AEROHEATING(PRESSURE) CHARACTERISTICS OF A 0.010-SCALE VERSION OF THE VEHICLE 3 SPACE SHUTTLE CONFIGURATION(26-OTS) IN THE LANGLEY RESEARCH CENTER 4-FOOT WIND TUNNEL(IH4)
2138	144,610	V-03	IH4	AEROHEATING(PRESSURE) CHARACTERISTICS OF A 0.010-SCALE VERSION OF THE VEHICLE 3 SPACE SHUTTLE CONFIGURATION(26-OTS) IN THE LANGLEY RESEARCH CENTER 4-FOOT WIND TUNNEL(IH4)
2138	144,611	V-04	IH4	AEROHEATING(PRESSURE) CHARACTERISTICS OF A 0.010-SCALE VERSION OF THE VEHICLE 3 SPACE SHUTTLE CONFIGURATION(26-OTS) IN THE LANGLEY RESEARCH CENTER 4-FOOT WIND TUNNEL(IH4)
2139	134,407		OA118	EFFECT OF ELEVON GAP CONFIGURATIONS ON THE LONGITUDINAL AND LATERAL/DIRECTIONAL STABILITY AND CONTROL EFFECTIVENESS OF THE 43-0 SPACE SHUTTLE ORBITER (IA60/OA105)
2140	134,408		OA37	INVESTIGATION OF SPACE SHUTTLE ORBITER SUBSONIC STABILITY AND CONTROL CHARACTERISTICS AND DETERMINATION OF CONTROL SURFACE HINGE MOMENTS IN THE ROCKWELL INTERNATIONAL LOW SPEED WIND TUNNEL (OA37)

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2141	141,538		OH11	RESULTS OF TESTS OF A ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER (-139 CONFIGURATION) 0.0175-SCALE MODEL (NO.29-O) IN THE AEDC TUNNEL.F TO DETERMINE HYPERSONIC HEATING EFFECTS (OH11)
2142	134,402		FA4	DETERMINATION OF AERODYNAMIC STABILITY AND DRAG OF THE TITAN SRM DURING ENTRY
2143	144,587		IA61A	AERODYNAMIC RESULTS OF WIND TUNNEL TESTS ON AN 0.010-SCALE MODEL (32-OTS) SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC VKF 40-INCH SUPERSONIC WIND TUNNEL
2144	134,427		IA68	AN INVESTIGATION OF THE SUPPORT INTERFERENCE EFFECTS OF THE SSV MODEL 13P-OTS IN THE TRANSONIC AND SUPERSONIC FLOW REGIMES
2145	134,420		TA1F	AN INVESTIGATION TO DETERMINE THE STATIC STABILITY DURING RE-ENTRY OF THE 0.003-SCALE MCR 0200 BASELINE SPACE SHUTTLE EXTERNAL TANK MODEL
2146	134,092		IS4	FLUTTER TESTS (IS4) OF THE 0.0125-SCALE SHUTTLE REFLECTION PLANE MODEL 30-OTS IN THE LANGLEY RESEARCH CENTER 26-INCH TRANSONIC BLOWDOWN TUNNEL TEST NO. 547
2147	134,097		OA20C	RESULTS OF INVESTIGATIONS (OA20C) ON AN 0.015-SCALE CONFIGURATION 140A/B SPACE SHUTTLE VEHICLE ORBITER MODEL (49-O) IN THE NASA/LANGLEY RESEARCH CENTER UNITARY PLAN WIND TUNNEL
2148	134,440	V-01	IH20	HYPERSONIC AEROHEATING TEST OF SPACE SHUTTLE VEHICLE CONFIGURATION 3 (MODEL 22-OTS) IN THE NASA-AMES 3.5-FOOT HYPERSONIC WIND TUNNEL(IH-20)
2148	134,441	V-02	IH20	HYPERSONIC AEROHEATING TEST OF SPACE SHUTTLE VEHICLE CONFIGURATION 3 (MODEL 22-OTS) IN THE NASA-AMES 3.5-FOOT HYPERSONIC WIND TUNNEL(IH-20)
2149	141,805		OA90	RESULTS OF INVESTIGATIONS ON A 0.010-SCALE 140A/B CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 72-O IN THE NASA/LANGLEY RESEARCH CENTER CONTINUOUS FLOW HYPERSONIC TUNNEL (OA90)
2150	141,511		SA25F	AN INVESTIGATION OF HIGH MACH NUMBER STATIC STABILITY CHARACTERISTICS FOR A LARGE SCALE SOLID ROCKET BOOSTER
2151	141,815		OH6	RESULTS OF AERODYNAMIC HEAT TRANSFER TESTS OF A 0.0175-SCALE MODEL OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER 139 (MODEL NUMBER 22-O) IN THE NASA/AMES 3.5-FOOT HYPERSONIC WIND TUNNEL (TEST OH6)

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2152 R-01	134,423		OA81	RESULTS OF AN INVESTIGATION OF HYPERSONIC VISCOUS INTERACTION EFFECTS ON AN 0.01 SCALE SPACE SHUTTLE ORBITER S1-O MODEL IN THE AEDC-VKF HYPERVELOCITY WIND TUNNEL
2153	151,377		IH1	INVESTIGATION OF THE HEAT TRANSFER EFFECTS ON THE 22-OTS 0.0175- SCALE THIN SKIN THERMOCOUPLE MODEL (VEHICLE 3 CONFIGURATION)
2154	134,437		OH4A	HEAT TRANSFER TESTS OF A 0.0175-SCALE SPACE SHUTTLE ORBITER MODEL (29-O) TO DETERMINE THE EFFECT OF SURFACE TEMPERATURE ON BOUNDARY LAYER TRANSITION AT MACH 8.0 IN THE AEDC VKF TUNNEL B (TEST OH4A)
2155	134,406		OA110	STABILITY AND CONTROL CHARACTERISTICS FOR THE INNER MOLD LINE CONFIGURATION OF SPACE SHUTTLE ORBITER(OA110)
2156	141,797	V-01	IA17A	RESULTS OF AN EXTERNAL TANK SEPARATION TEST IN THE AEDC/VKF TUNNEL BON AN 0.010 SCALE REPLICA OF THE SPACE SHUTTLE VEHICLE (MODEL 52-OT)IA17A
2156	141,798	V-02	IA17A	RESULTS OF AN EXTERNAL TANK SEPARATING TEST IN THE AEDC/VKF TUNNEL BON AN 0.010 SCALE REPLICA OF THE SPACE SHUTTLE VEHICLE (MODEL 52-OT)IA17A
2156	141,799	V-03	IA17A	RESULTS OF AN EXTERNAL TANK SEPARATION TEST IN THE AEDC/VKF TUNNEL BON AN 0.010 SCALE REPLICA OF THE SPACE SHUTTLE VEHICLE (MODEL 52-OT)IA17A
2157	141,822		IH19	HEAT TRANSFER TESTS OF AN 0.006-SCALE THIN SKIN SPACE SHUTTLE MODEL (50-0. 41-T) IN THE LANGLEY RESEARCH CENTER NITROGEN TUNNEL AT MACH 19
2158	147,640		IS6A	FLOW VISUALIZATION TESTS OF A 0.004-SCALE SPACE SHUTTLE VEHICLE 2A MODEL (NO. 13-OTS) IN THE MSFC 14-INCH TRISONIC WIND TUNNEL
2159	134,410	V-01	OA59	AERODYNAMIC RESULTS OF SUPPORT SYSTEM EFFECTS TESTS CONDUCTED IN NASA/ARC 6-BY 6FOOT SUPERSONIC WIND TUNNEL USING A 0.015-SCALE MODEL OF THE CONFIGURATION 140A/B SSV ORBITER (OA59)
2159	134,412	V-02	OA59	AERODYNAMIC RESULTS OF SUPPORT SYSTEM EFFECTS TESTS CONDUCTED IN NASA/ARC 6-BY-6 FOOT SUPERSONIC WIND TUNNEL USING A 0.015 -SCALE MODEL OF THE CONFIGURATION 140A/B SSV ORBITER (OA59)
2160	134,413		IA18	WIND TUNNEL TESTS OF THE 0.010-SCALE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/AMES 3.5 FOOT HYPERSONIC WIND TUNNEL (IA18)
2161	134,422		SA6F	AERODYNAMIC CHARACTERISTICS OF MSFC MODEL 454 OF THE 142 INCH SOLID ROCKET BOOSTER TESTED IN THE LRPC 10-FOOT SWT AT MACH NUMBERS OF 2.0 AND 2.7 (SA6F)

CHRYSLER DATA MANAGEMENT SERVICES(DMS)
SPACE SHUTTLE WIND TUNNEL TEST PROGRAM
DATA REPORT DOCUMENTATION

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2162	134,430		OA36	RESULTS OF INVESTIGATIONS ON AN 0.015-SCALE 140A/B CONFIGURATION OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER IN THE NASA/AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL (OA36)
2163	134,403		OA208	AERODYNAMIC RESULTS OF A SUPPORT SYSTEM INTERFERENCE EFFECTS TEST CONDUCTED AT NASA/LARC UPWT USING AN 0.015-SCALE MODEL OF THE CONFIGURATION 140A/B SSV ORBITER (OA208)
2164	141,828	V-01	OH12/IH21	HEAT TRANSFER TESTS ON A 0.01-SCALE ROCKWELL CONFIGURATION 3 SPACE SHUTTLE ORBITER AND TANK (37-OT)IN THE CALSPAN 48-INCH HYPERSONIC SH OCK TUNNEL (OH12/IH21)
2164	141,829	V-02	OH12/IH21	HEAT TRANSFER TESTS ON A 0.01-SCALE ROCKWELL CONFIGURATION 3 SPACE SHUTTLE ORBITER AND TANK (37-OT)IN THE CALSPAN 48-INCH HYPERSONIC SH OCK TUNNEL (OH12/IH21)
2164	141,830	V-03	OH12/IH21	HEAT TRANSFER TESTS ON A 0.01-SCALE ROCKWELL CONFIGURATION 3 SPACE SHUTTLE ORBITER AND TANK (37-OT)IN THE CALSPAN 48-INCH HYPERSONIC SH OCK TUNNEL (OH12/IH21)
2165	141,823	V-01	TA2F	RESULTS OF AN INVESTIGATION OF AN 0.003-SCALE SPACE SHUTTLE EXTERNALTANK (MSFC MODEL 460) IN THE NASA/MSFC 14 X 14-INCH TRISONIC WIND TUNNEL TO DETERMINE STATIC PRESSURE DISTRIBUTIONS DURING REENTRY (TA2F)
2165	141,824	V-02	TA2F	RESULTS OF AN INVESTIGATION OF AN 0.003-SCALE SPACE SHUTTLE EXTERNALTANK (MSFC MODEL 460) IN THE NASA/MSFC 14 X 14-INCH TRISONIC WIND TUNNEL TO DETERMINE STATIC PRESSURE DISTRIBUTIONS DURING REENTRY (TA2F)
2165	141,825	V-03	TA2F	RESULTS OF AN INVESTIGATION OF AN 0.003-SCALE SPACE SHUTTLE EXTERNALTANK (MSFC MODEL 460) IN THE NASA/MSFC 14 X 14-INCH TRISONIC WIND TUNNEL TO DETERMINE STATIC PRESSURE DISTRIBUTIONS DURING REENTRY (TA2F)
2165	141,826	V-04	TA2F	RESULTS OF AN INVESTIGATION OF AN 0.003-SCALE SPACE SHUTTLE EXTERNALTANK (MSFC MODEL 460) IN THE NASA/MSFC 14 X 14-INCH TRISONIC WIND TUNNEL TO DETERMINE STATIC PRESSURE DISTRIBUTIONS DURING REENTRY (TA2F)
2165	141,827	V-05	TA2F	RESULTS OF AN INVESTIGATION OF AN 0.003-SCALE SPACE SHUTTLE EXTERNALTANK (MSFC MODEL 460) IN THE NASA/MSFC 14 X 14-INCH TRISONIC WIND TUNNEL TO DETERMINE STATIC PRESSURE DISTRIBUTIONS DURING REENTRY (TA2F)

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2166	141,534		IH16	HEAT TRANSFER TESTS OF AN 0.006 SCALE THIN-SKIN SPACE SHUTTLE THERMOCOUPLE MODEL (41-OTS) IN THE LANGLEY RESEARCH CENTER UNITARY PLAN WIND TUNNEL AT M=3.7 (IH16)
2167	141,550		OA98	RESULTS OF AN INVESTIGATION ON AN 0.015-SCALE MODEL(49-O) OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER IN THE NASA AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL (OA98)
2168	TM-X71945		LA32	HEAT TRANSFER TO SURFACE AND GAPS OF RSI TILE ARRAYS IN TURBULENT FLOW AT MACH 10.3
2169	141,836	V-01	IA81A	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 11 X 11 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81A) VOLUME 1 OF 7
2169	141,837	V-02	IA81A	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 11 X 11 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81A) VOLUME 2 OF 7
2169	141,838	V-03	IA81A	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 11 X 11 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81A) VOLUME 3 OF 7
2169	141,839	V-04	IA81A	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 11 X 11 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81A) VOLUME 4 OF 7
2169	141,840	V-05	IA81A	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 11 X 11 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81A) VOLUME 5 OF 7
2169	141,841	V-06	IA81A	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 11 X 11 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81A) VOLUME 6 OF 7
2169	141,842	V-07	IA81A	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 11 X 11 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81A) VOLUME 7 OF 7

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2170	141,543	V-01	IA19	RESULTS OF A JET PLUME EFFECTS TEST ON THE ROCKWELL INTERNATIONAL INTEGRATED SPACE SHUTTLE VEHICLE USING A VEHICLE 5 CONFIGURATION 0.02-SCALE MODEL (88-OTS) IN THE 11 X 11 FOOT LEG OF THE NASA/AMES RESEARCH CENTER UNITARY PLAN WIND TUNNEL (IA19)
2170	141,544	V-02	IA19	RESULTS OF A JET PLUME EFFECTS TEST ON THE ROCKWELL INTERNATIONAL INTEGRATED SPACE SHUTTLE VEHICLE USING A VEHICLE 5 CONFIGURATION 0.02-SCALE MODEL (88-OTS) IN THE 11 X 11 FOOT LEG OF THE NASA/AMES RESEARCH CENTER UNITARY PLAN WIND TUNNEL (IA19)
2170	141,545	V-03	IA19	RESULTS OF A JET PLUME EFFECTS TEST ON THE ROCKWELL INTERNATIONAL INTEGRATED SPACE SHUTTLE VEHICLE USING A VEHICLE 5 CONFIGURATION 0.02-SCALE MODEL (88-OTS) IN THE 11 X 11 FOOT LEG OF THE NASA/AMES RESEARCH CENTER UNITARY PLAN WIND TUNNEL (IA19)
2171	144,584	V-01	OH38	RESULTS OF PRESSURE DISTRIBUTION TESTS OF A 0.010-SCALE SPACE SHUTTLE ORBITER MODEL (61-0) IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (OH38)
2171	144,585	V-02	OH38	RESULTS OF PRESSURE DISTRIBUTION TESTS OF A 0.010-SCALE SPACE SHUTTLE ORBITER MODEL (61-0) IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (OH38)
2171	144,586	V-03	OH38	RESULTS OF PRESSURE DISTRIBUTION TES S OF A 0.010-SCALE SPACE SHUTTLE ORBITER MODEL (61-0) IN THE NASA/ARC 3.5-FO HYPERSONIC WIND TUNNEL (OH38)
2172	134,415		OA99	RESULTS OF REACTION CONTROL SYSTEM ON-ORBIT JET USING AN 0.0175-SCALE CONFIGURATION 3 SPACE SHUTTLE ORBITER MODEL (21-0) IN THE LARC 60-FOOT VACUUM SPHERE
2173	134,107		IA8	AERODYNAMIC RESULTS OF AN ABORT SEPARATION EFFECTS TEST (IA8) CONDUCTED IN THE NASA/LARC 14-FOOT TRANSONIC WIND TUNNEL ON A MODEL (6-OTS) OF THE ROCKWELL INTERNATIONAL LAUNCH CONFIGURATION INTEGRATED VEHICLE
2174	141,811	V-01	IA33	AN INVESTIGATION IN THE MSFC 14-INCH TWT TO DETERMINE THE STATIC STABILITY CHARACTERISTICS OF THE 0.004-SCALE MODEL (74-OTS) SPACE SHUTTLE VEHICLE 5 CONFIGURATION (IA33)
2174	141,812	V-02	IA33	AN INVESTIGATION IN THE MSFC 14-INCH TWT TO DETERMINE THE STATIC STABILITY CHARACTERISTICS OF THE 0.004-SCALE MODEL (74-OTS) SPACE SHUTTLE VEHICLE 5 CONFIGURATION (IA33)

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2174	141,813	V-03	IA33	AN INVESTIGATION IN THE MSFC 14-INCH TWT TO DETERMINE THE STATIC STABILITY CHARACTERISTICS OF THE 0.004-SCALE MODEL (74-OTS) SPACE SHUTTLE VEHICLE 5 CONFIGURATION (IA33)
2175	134,431	V-01	IA70	SUBSONIC AND TRANSONIC HINGE MOMENT AND WING BENDING/TORSION CHARACTERISTICS FOR THE -140A/B INTEGRATED SPACE SHUTTLE VEHICLE (IA70) VOLUME 1 OF 3
2175	134,432	V-02	IA70	SUBSONIC AND TRANSONIC HINGE MOMENT AND WING BENDING/TORSION CHARACTERISTICS FOR THE -140A/B INTEGRATED SPACE SHUTTLE VEHICLE (IA70) VOLUME 2 OF 3
2175	134,433	V-03	IA70	SUBSONIC AND TRANSONIC HINGE MOMENT AND WING BENDING/TORSION CHARACTERISTICS FOR THE -140A/B INTEGRATED SPACE SHUTTLE VEHICLE (IA70) VOLUME 3 OF 3
2176	TM-X72661		LA40	SPACE SHUTTLE ORBITER TRIMMED CENTER OF GRAVITY EXTENSION STUDY VOLUME IV - EFFECTS OF CONFIGURATION MODIFICATIONS ON THE AERODYNAMICS OF THE 139B ORBITER AT MACH 20.3
2177	141,510		OA83	RESULTS OF INVESTIGATIONS ON AN 0.015-SCALE CONFIGURATION 140A/B SPACE SHUTTLE VEHICLE ORBITER REACTION CONTROL SYSTEM PLUME-IMPINGEMENT MODEL 36-0 IN THE NASA/AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL (OA83)
2178	134,119		OA53B	INVESTIGATIONS ON AN 0.030-SCALE SPACE SHUTTLE VEHICLE CONFIGURATION140A/B ORBITER MODEL IN THE AMES RESEARCH CENTER 9- BY 7-FOOT SUPER-SONIC WIND TUNNEL (OA53B)
2179	151,378		OS8A/B	RESULTS OF AN INVESTIGATION OF THE ACOUSTIC AND VIBRATIONAL ENVIRONMENT OF A FULL SCALE SPACE SHUTTLE ORBITER STRUCTURAL TEST PANEL WITHSIMULATED TPS IN THE AMES UNITARY PLAN WIND TUNNEL (MODEL 81-0.TEST OS8A AND B)
2180	147,615	V-01	IH28	HEAT TRANSFER TEST OF AN 0.006-SCALE THIN-SKIN THERMOCOUPLE SPACE SHUTTLE MODEL (50-0.41T) IN THE NASA-AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL AT MACH 5.3 (IH-28)
2180	147,616	V-02	IH28	HEAT TRANSFER TEST OF AN 0.006-SCALE THIN-SKIN THERMOCOUPLE SPACE SHUTTLE MODEL (50-0.41T) IN THE NASA-AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL AT MACH 5.3 (IH-28)
2181	134,425		TA9F	A HYPERSONIC FORCE AND MOMENT TEST OF A 0.006 SCALE MODEL OF THE 330.2 INCH DIAMETER EXTERNAL TANK IN THE AMES RESEARCH CENTER 3.5 FT. HYPERSONIC WIND TUNNEL (TA9F)

DMS-DR REPORT NUMBER	NASA CR NUMBER	NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2182	151,062	LA49	SUPERSONIC CONTROL EFFECTIVENESS FOR FULL AND PARTIAL SPAN ELEVON CONFIGURATIONS ON A 0.0165 SCALE MODEL SPACE SHUTTLE ORBITER TESTED IN THE LARC UNITARY PLAN WIND TUNNEL
2183	TM-X72661	LA51	SPACE SHUTTLE ORBITER TRIMMED CENTER-OF-GRAVITY EXTENSION STUDY: VOLUME II-EFFECTS OF CONFIGURATION MODIFICATIONS ON THE AERODYNAMIC CHARACTERISTICS OF THE 140A/B ORBITER AT TRANSONIC SPEEDS
2184	151,061	LA48	TRANSONIC CONTROL EFFECTIVENESS FOR FULL AND PARTIAL SPAN ELEVON CONFIGURATIONS ON A 0.0165 SCALE MODEL SPACE SHUTTLE ORBITER TESTED INTHE LARC 8-FOOT TRANSONIC PRESSURE TUNNEL
2185	134,120	OA53C	INVESTIGATIONS ON AN 0.030-SCALE SPACE SHUTTLE VEHICLE CONFIGURATION140A/B ORBITER MODEL IN THE AMES RESEARCH CENTER UNITARY PLAN 8-BY 7-FOOT SUPERSONIC WIND TUNNEL
2186	134,428	OA116	RESULTS OF DIFFERENTIAL ELEVON/AILERON DEFLECTION FOR LATERAL CONTROL OPTIMIZATION AND ELEVON HINGE MOMENT INVESTIGATIONS ON AN 0.015-SCALE MODEL(49-0) OF THE SPACE SHUTTLE ORBITER IN THE NASA/LANGLEY RESEARCH CENTER 8-FOOT TRANSONIC PRESSURE TUNNEL
2187	134,421	OA119A	EFFECTS OF WING/ELEVON GAP SEALING FLAPPER DOORS ON ORBITER ELEVON EFFECTIVENESS (OA119A)
2188	TM-X	LA39	** TO BE PUBLISHED AT LARC **
2189	141,506	IA110	RESULTS OF INVESTIGATION IA110 ON A 0.015-SCALE INTEGRATED CONFIGURATION OF THE SPACE SHUTTLE VEHICLE IN THE ARC 9X7 SUPERSONIC WIND TUNNEL USING MODELS 67-TS AND 49-0
2190	141,537	OA108	INVESTIGATION IN THE MSFC TWT TO VERIFY THE STATIC STABILITY AND CONTROL EFFECTIVENESS OF THE 0.004-SCALE MODEL (74-0) OF THE SHUTTLE 5 ORBITER (OA-108)
2191	TM-X72661	LA47	SPACE SHUTTLE ORBITER TRIMMED CENTER OF GRAVITY EXTENSION STUDY: VOLUME 1--EFFECTS OF CONFIGURATIONS ON THE AERODYNAMIC CHARACTERISTICS OF THE 140 A/B ORBITER AT MACH 10.3
2192	141,541 V-01	IA87	AERODYNAMIC RESULTS OF A SEPARATION EFFECTS TEST (IA87) ON A 0.01-SCALE MODEL (52-OTS) OF THE INTEGRATED SSV IN THE AEDC/VKF 40-BY-40 INCH SUPERSONIC WIND TUNNEL A

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2192	141,542	V-02	IA87	AERODYNAMIC RESULTS OF A SEPARATION EFFECTS TEST (IA87) ON A 0.01-SCALE MODEL (52-OTS) OF THE INTEGRATED SSV IN THE AEDC/VKF 40-BY-40 INCH SUPERSONIC WIND TUNNEL A
2193	151,380		OH26	RESULTS OF HEAT TRANSFER TEST OF A 0.0175-SCALE SPACE SHUTTLE ORBITER 140B MODEL (MODIFIED 22-O) IN THE NASA-AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL
2194	141,817	V-01	IA81B	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 9 X 7 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81B) VOLUME 1 OF 5
2194	141,818	V-02	IA81B	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 9 X 7 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81B) VOLUME 2 OF 5
2194	141,819	V-03	IA81B	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 9 X 7 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81B) VOLUME 3 OF 5
2194	141,820	V-04	IA81B	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 9 X 7 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81B) OVLUME 4 OF 5
2194	141,821	V-05	IA81B	RESULTS OF A PRESSURE LOADS INVESTIGATION ON A 0.030-SCALE MODEL (47-OTS) OF THE INTEGRATED SPACE SHUTTLE VEHICLE CONFIGURATION 5 IN THENASA AMES RESEARCH CENTER 9 X 7 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL (IA81B) VOLUME 5 OF 5
2195	134,442		OA82	RESULTS OF TEST OA82 IN THE NASA/LRC 31-INCH CFHT ON AN 0.010-SCALE MODEL(32-O) OF THE SPACE SHUTTLE CONFIGURATION 3 TO DETERMINE RCS JET FLOW FIELD INTERACTION AND TO INVESTIGATE RT REAL GAS EFFECTS
2196	141,531		OA79	RESULTS OF INVESTIGATIONS OF AN 0.015 SCALE SPACE SHUTTLE VEHICLE 140A/B CONFIGURATION WITH MODIFIED OMS PODS AND ELEVONS IN THE AEDC VKF TUNNEL B (OA79)
2197	134,418		FH10	PRESSURE AND HEAT-FLUX RESULTS FROM THE SPACE SHUTTLE/EXTERNAL FUEL TANK INTERACTION TEST AT MACH NUMBERS 16 AND 19 (FH10)
2198	141,534		OA115	DIFFERENTIAL ELEVON EFFECTIVENESS LATERAL CONTROL OPTIMIZATION AND ELEVON HINGE MOMENT INVESTIGATION ON A 0.015-SCALE SPACE SHUTTLE ORBITER MODEL (140 A/B/C MODIFIED) IN THE AEDC VKF WIND TUNNEL A (OA115)

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2199	TM-X3315		LA43A/B/LA43B	SUPERSONIC DYNAMIC-STABILITY DERIVATIVES OF THE SPACE SHUTTLE LAUNCHVEHICLE
2200	TM-X3336		LA44	SUBSONIC AND TRANSONIC DYNAMIC-STABILITY CHARACTERISTICS OF THE SPACE SHUTTLE LAUNCH VEHICLE
2201	160,854		CA3	MATED CARRIER AERODYNAMIC CHARACTERISTICS INVESTIGATION FOR 0.04-SCALE MODEL BOEING 747 CARRIER (MODEL TE 1065)/SS ORBITER (MODEL 43-O) AND 747 CARRIER/ET (MODEL 1284-72) COMBINATIONS IN THE U. OF WASH. AERONAUTICAL LABORATORY (UWAL) F.K.KIRSTEN WIND TUNNEL (CA3)
2202	141,526		OA123	SPACE SHUTTLE VEHICLE FERRY CONFIGURATION AFTERBODY FAIRING EFFECTS ON 140A/B ORBITER AERODYNAMIC CHARACTERISTICS USING AN .0405-SCALE MODEL ORBITER (43-O) IN THE ROCKWELL INTERNATIONAL 7.75 X 11 FT LOW SPEED WIND TUNNEL (OA123)
2203	141,524		OA119B	RESULTS OF AN INVESTIGATION OF ELEVON HINGE MOMENTS AND DUAL PANEL ELEVON EFFECTIVENESS USING AN .0405-SCALE MODEL (16-O) OF THE CONFIGURATION 140C SPACE SHUTTLE ORBITER IN THE ROCKWELL INTERNATIONAL NAALLOW SPEED WIND TUNNEL (OA119B)
2204	141,525		IA43	RESULTS OF TRANSONIC WIND TUNNEL TESTS ON AN 0.010-SCALE SPACE SHUTTLE MATED VEHICLE MODEL 72-OTS IN THE LARC 8-FOOT TPT (IA43)
2205	141,532		OA109	RESULTS OF A 0.004-SCALE 140C MODIFIED CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL (74-O) IN THE NASA/LANGLEY RESEARCH CENTER HYPERSONIC HELIUM TUNNEL (OA109)
2206	141,528		IA44	RESULTS OF INVESTIGATIONS ON AN 0.010-SCALE 140A/B CONFIGURATION (MODEL 72OTS) OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER IN THE NASA/LANGLEY RESEARCH CENTER UNITARY PLAN WIND TUNNEL (IA44)
2207	147,608		SA29F	AN INVESTIGATION TO DETERMINE THE PRESSURE DISTRIBUTION ON THE 0.0137 SCALE SOLID ROCKET BOOSTER FOREBODY (MSFC MODEL 467) AT HIGH ANGLES OF ATTACK AT OR NEAR 90 DEGREES AND HIGH REYNOLDS NUMBERS IN THE MSFC HIGH REYNOLDS NUMBER WIND TUNNEL
2208	144,590	V-01	TA3F	AN INVESTIGATION OF THE 0.0091SCALE EXTERNAL TANK OGIVE NOSE (MSFC MODEL 470) IN THE MSFC 14 INCH TWT TO DETERMINE THE PRESSURE DISTRIBUTION AROUND THE EXTERNAL TANK NOSE
2208	144,591	V-02	TA3F	AN INVESTIGATION OF THE 0.0091SCALE EXTERNAL TANK OGIVE NOSE (MSFC MODEL 470) IN THE MSFC 14 INCH TWT TO DETERMINE THE PRESSURE DISTRIBUTION AROUND THE EXTERNAL TANK NOSE

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2209	141,536		OA124	RESULTS OF A SPACE SHUTTLE VEHICLE FERRY CONFIGURATION AFTERBODY FAIRING OPTIMIZATION STUDY USING A 140A/B 0.0405-SCALE MODEL ORBITER (43-0) IN THE ROCKWELL INTERNATIONAL 7.75 X 11.0 FT LOW SPEED WIND TUNNEL (OA124)
2210	151,372		IH27	CONNECTIVE HEAT-TRANSFER TEST RESULTS FOR A GAP, CYLINDRICAL-PROTUBERANCE, AND SHOCK-IMPINGEMENT FLAT-PLATE MODEL IN THE NASA-AMES 3.5-FOOT HYPERSONIC WIND TUNNEL (TEST IH27, MODEL 15-0 VIII)
2211	141,800	V-01	CA5	RESULTS OF A 0.03-SCALE AERODYNAMIC CHARACTERISTICS INVESTIGATION OFA BOEING 747 CARRIER(MODEL NO. AX 1319 I-1) MATED WITH A SPACE SHUTTLE ORBITER (MODEL 45-0) CONDUCTED IN THE BOEING TRANSONIC WIND TUNNEL (CA5)
2211	141,803	V-02	CA5	RESULTS OF A 0.03-SCALE AERODYNAMIC CHARACTERISTICS INVESTIGATION OF LE ORBITER (MODEL 45-0) CONDUCTED IN THE BOEING TRANSONIC WIND TUNNEL (CA5)
2211	141,804	V-03	CA5	RESULTS OF A 0.03-SCALE AERODYNAMIC CHARACTERISTICS INVESTIGATION OFA BOEING 747 CARRIER(MODEL NO. AX-1319 I-1) MATED WITH A SPACE SHUTTLE ORBITER (MODEL 45-0) CONDUCTED IN THE BOEING TRANSONIC WIND TUNNEL (CA5)
2212	147,632	V-01	IA80	INVESTIGATIONS OF THE 0.020-SCALE 88-OTS INTEGRATED SPACE SHUTTLE VEHICLE JET-PLUME MODEL IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT UNITARY PLAN WIND TUNNEL (IA80)
2212	147,633	V-02	IA80	INVESTIGATIONS OF THE 0.020-SCALE 88-OTS INTEGRATED SPACE SHUTTLE VEHICLE JET-PLUME MODEL IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT UNITARY PLAN WIND TUNNEL (IA80)
2212	147,634	V-03	IA80	INVESTIGATIONS OF THE 0.020-SCALE 88-OTS INTEGRATED SPACE SHUTTLE VEHICLE JET-PLUME MODEL IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT UNITARY PLAN WIND TUNNEL (IA80)
2212	147,635	V-04	IA80	INVESTIGATIONS OF THE 0.020-SCALE 88-OTS INTEGRATED SPACE SHUTTLE VEHICLE JET-PLUME MODEL IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT UNITARY PLAN WIND TUNNEL (IA80)
2213			LA53/LA54	** DOCUMENTATION NOT COMPLETE **
2214	141,513		OA89	RESULTS OF INVESTIGATIONS ON AM 0.004-SCALE 140C MODIFIED CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL (74-0) IN THE NASA/LANGLEY RESEARCH CENTER HYPERSONIC NITROGEN TUNNFL (OA89)

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2215	144,592		LA58	UPPER WING SURFACE BOUNDARY LAYER MEASUREMENTS AND STATIC AERODYNAMIC DATA OBTAINED ON AN 0.015-SCALE MODEL OF THE SSV ORBITER CONFIGURATION 140A/B IN THE LTV ASWT AT A MACH NUMBER OF 4.6 (LA58)
2216	141,802		SH12F	RESULTS OF AEROTHERMODYNAMIC HEATING TEST ON A 0.013 SCALE MODEL SOLID ROCKET BOOSTER IN THE NASA/LARC UNITARY PLAN WIND TUNNEL (SH12F)
2217	141,844	V-01	CA20	AERODYNAMIC RESULTS OF A SEPARATION TEST(CA20) CONDUCTED AT THE BOEING TRANSONIC WIND TUNNEL USING 0.030-SCALE MODELS OF THE CONFIGURATION 140A/B (MODIFIED) SSV ORBITER (MODEL NO. 45-0) AND THE BEOING 747CARRIER (MODEL NO. AX 1319 I-1)
2217	141,845	V-02	CA20	AERODYNAMIC RESULTS OF A SEPARATION TEST(CA20) CONDUCTED AT THE BOEING TRANSONIC WIND TUNNEL USING 0.030-SCALE MODELS OF THE CONFIGURATION 140A/B (MODIFIED) SSV ORBITER (MODEL NO. 45-0) AND THE BEOING 747CARRIER (MODEL NO. AX 1319 I-1)
2217	141,846	V-03	CA20	AERODYNAMIC RESULTS OF A SEPARATION TEST(CA20) CONDUCTED AT THE BOEING TRANSONIC WIND TUNNEL USING 0.030-SCALE MODELS OF THE CONFIGURATION 140A/B (MODIFIED) SSV ORBITER (MODEL NO. 45-0) AND THE BEOING 747CARRIER (MODEL NO. AX 1319 I-1)
2218	151,367		TH1F	PRESSURE AND HEAT TRANSFER TESTS RESULTS ON THE SPACE SHUTTLE 0.015-SCALE EXTERNAL TANK AT MACH 16 IN AEDC TUNNEL F
2219	144,597	V-01	IA82C	RESULTS OF AN INVESTIGATION OF JET PLUME EFFECTS ON AN 0.010-SCALE MODEL (75-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE 8- BY 7-FOOT LEG OF THE NASA/AMES UNITARY WIND TUNNEL (IA82C)
2219	144,598	V-02	IA82C	RESULTS OF AN INVESTIGATION OF JET PLUME EFFECTS ON AN 0.010-SCALE MODEL (75-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE 8- BY 7-FOOT LEG OF THE NASA/AMES UNITARY WIND TUNNEL (IA82C)
2220	TM-X		LA52	** TO BE PUBLISHED AT LARC **
2221	141,548		OA143	INVESTIGATION OF SPACE SHUTTLE VEHICLE 140C CONFIGURATION ORBITER (MODEL 16-0) WHEEL WELL PRESSURE LOADS IN THE ROCKWELL INTERNATIONAL 7.75 X 11 FOOT WIND TUNNEL (OA143)
2222	147,626	V-01	OH49B	RESULTS FROM A CONVECTIVE HEAT-TRANSFER-RATE DISTRIBUTION TEST ON A 0.0175 SCALE MODEL(22-0) OF THE ROCKWELL INTERNATIONAL VEHICLE 4 SPACE SHUTTLE CONFIGURATION IN THE AEDC-VKF TUNNEL B(OH49B)

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2222	147,627	V-02	OH49B	RESULTS FROM A CONVECTIVE HEAT-TRANSFER-RATE DISTRIBUTION TEST ON A 0.0175 SCALE MODEL(22-0) OF THE ROCKWELL INTERNATIONAL VEHICLE 4 SPACE SHUTTLE CONFIGURATION IN THE AEDC-VKF TUNNEL B(OH49B)
2223	141,549		SA8F	REENTRY STATIC STABILITY CHARACTERISTICS OF A .005479 SCALE MODEL 146-INCH SOLID ROCKET BOOSTER TESTED IN THE NASA/MSFC 14X14 INCH TWT
2224	147,650		LA56	RESULTS OF A DRAG REDUCTION INVESTIGATED ON AN 0.010-SCALE MODEL OF THE SPACE SHUTTLE VEHICLE 72-OTS LAUNCH CONFIGURATION TESTED IN THE LARC 8-FOOT TRANSONIC PRESSURE TUNNEL FOR THE MACH RANGE OF 0.35 TO 1.20 (LA56)
2225	141,505		OH4C	PHASE CHANGE PAINT TESTS TO INVESTIGATE EFFECTS OF TPS TILES ON HEATING RATES OF THE ROCKWELL SPACE SHUTTLE ORBITER (TEST OH4C, MODEL 21-0)
2226	141,507		IA61B	RESULTS OF FLOW VISUALIZATION TESTS OF 0.010-SCALE SPACE SHUTTLE MODELS 32-OTS AND 52-0 IN THE AEDC VKF TUNNEL A (IA61B)
2227	141,806		IA71	RESULTS OF EXPERIMENTAL TESTS IN THE MSFC 14X14 INCH TRISONIC TUNNELON A .004 SCALE MODEL SPACE SHUTTLE INTEGRATED VEHICLE 5 (MODEL 77-0, 74-TS) TO RELIEVE WING LOADS DURING ASCENT (IA71)
2228	TM-X72661		LA46A/B	** TO BE PUBLISHED AT LARC **
2229	141,508		OA102	RESULTS OF FLOW-VISUALIZATION INVESTIGATIONS ON A 0.015-SCALE MODIFIED CONFIGURATION 140A/B SPACE SHUTTLE VEHICLE ORBITER (MODEL 36-0) IN THE LANGLEY RESEARCH CENTER
2230	141,509		IA17B	RESULTS OF OIL FLOW VISUALIZATIONS TESTS OF AN 0.010-SCALE MODEL (52-OT) OF THE SPACE SHUTTLE ORBITER-TANK MATED AND ORBITER CONFIGURATIONS IN THE AEDC VKF TUNNEL B (IA17B)
2231	144,601	V-01	IA82B	RESULTS OF AN INVESTIGATION OF JET PLUME EFFECTS ON AN 0.010-SCALE MODEL (75-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE 9- BY 7-FOOT LEG OF THE NASA/AMES UNITARY WIND TUNNEL (IA82C)
2231	144,602	V-02	IA82B	RESULTS OF AN INVESTIGATION OF JET PLUME EFFECTS ON AN 0.010-SCALE MODEL (75-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE 9- BY 7-FOOT LEG OF THE NASA/AMES UNITARY WIND TUNNEL (IA82C)
2232	141,521		OA131	RESULTS OF INVESTIGATIONS ON THE 0.004-SCALE MODEL 74-0 OF THE CONFIGURATION 4 (MODIFIED) SPACE SHUTTLE VEHICLE ORBITER IN THE NASA/MSFC 14-BY-14-INCH TRISONIC WIND TUNNEL (OA131)

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2233	151,068		LA59	RESULTS OF A DRAG REDUCTION INVESTIGATION ON AN 0.010-SCALE MODEL OFTHE SPACE SHUTTLE VEHICLE 72-OTS LAUNCH CONFIGURATION TESTED IN THE LARC 8-FOOT TRANSONIC PRESSURE TUNNEL FOR THE MACH RANGE OF 0.3K TO 1.20
2234	141,547		OA113	WIND TUNNEL TEST OA113 OF THE 0.010-SCALE SPACE SHUTTLE ORBITER MODEL 51-O IN THE CALSPAN HYPERSONIC SHOCK TUNNEL (48-INCH LEG)
2235	141,810		SA30F	REENTRY AERODYNAMIC FORCES AND MOMENTS ON THE ENGINE NOZZLE OF THE 146-INCH SOLID ROCKET BOOSTER NODEL 473 IN MSFC 14 X 14 INCH TRISONICWIND TUNNEL (SA30F)
2236	141,835		CA11	MATED AERODYNAMIC CHARACTERISTICS INVESTIGATION FOR 0.04-SCALE MDOELBOEING 747 CAM/EXTERNAL TANK (MODEL AX1284 E-5) COMBINATION IN THE UNIVERSITY OF WASHINGTON AERONAUTICAL LABORATORY F.K. KIRSTEN WIND TUNNEL (CA11)
2237			OA155	** DOCUMENTATION NOT COMPLETE **
2238	141,847		OA93	RESULTS OF WIND TUNNEL RCS INTERACTION TESTS ON A 0.010-SCALE SPACE SHUTTLE ORBITER MODEL (51-O) IN THE CALSPAN CORPORATION 48-INCH HYPERSONIC SHOCK TUNNEL
2239	TM-X		LA38B	** TO BE PUBLISHED AT LARC **
2240	151,054		IH41A	RESULTS OF AN INVESTIGATION OF THE SPACE SHUTTLE INTEGRATED VEHICLE AERODYNAMIC HEATING CHARACTERISTICS OBTAINED USING THE 0.0175-SCALE MODEL 60-OTS IN THE AEDC TUNNEL A DURING TESTS IH41 AND IH41A
2241	160,490	V-01	OH39	AN INVESTIGATION OF ENTRY HEATING ON THE 0.0175 SCALE SPACE SHUTTLE ORBITER (MODEL 60-O) IN THE AEDC UKF TUNNEL B
2241	160,491	V-02	OH39	AN INVESTIGATION OF ENTRY HEATING ON THE 0.0175 SCALE SPACE SHUTTLE ORBITER (MODEL 60-O) IN THE AEDC UKF TUNNEL B
2241	160,492	V-03	OH39	AN INVESTIGATION OF ENTRY HEATING ON THE 0.0175 SCALE SPACE SHUTTLE ORBITER (MODEL 60-O) IN THE AEDC UKF TUNNEL B
2241	160,493	V-04	OH39	AN INVESTIGATION OF ENTRY HEATING ON THE 0.0175 SCALE SPACE SHUTTLE ORBITER (MODEL 60-O) IN THE AEDC UKF TUNNEL B
2242	141,831	V-01	IA111	AERODYNAMIC RESULTS OF A SEPARATION EFFECTS TEST ON A 0.010-SCALE MODEL (52-OTS) OF THE INTEGRATED SSV IN THE AEDC/VKF 40-BY-40 INCH SUPERSONIC WIND TUNNEL A (IA111)

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DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2242	144,588	V-02	IA111	AERODYNAMIC RESULTS OF A SEPARATION EFFECTS TEST ON A 0.010-SCALE MODEL (52-OTS) OF THE INTEGRATED SSV IN THE AEDC/VKF 40-BY-40 INCH SUPERSONIC WIND TUNNEL A (IA111)
2243	144,583		CA23A	RESULTS OF AN AERODYNAMIC INVESTIGATION OF A SPACE SHUTTLE ORBITER/747 CARRIER VEHICLE CONFIGURATION TO ESTABLISH A FREE-STREAM DATA BASE FOR ALT SEPARATION INVESTIGATIONS UTILIZING A 0.0125-SCALE MODEL (48-/OAX1318I-1) IN THE ARC 14-FOOT WIND TUNNEL (CA23A)
2244	151,082		SA28F	AN INVESTIGATION TO DETERMINE THE STATIC PRESSURE DISTRIBUTION OF THE 0.00548 SCALE SPACE SHUTTLE SOLID ROCKET BOOSTER (MSFC MODEL NUMBER 468) DURING REENTRY IN THE NASA/MSFC 14 INCH TRISONIC WIND TUNNEL
2245	147,618	V-01	OA161A/B/C	RESULTS OF AN INVESTIGATION TO DETERMINE LOCAL FLOW CHARACTERISTICS AT THE AIR DATA PROBE LOCATIONS USING AN 0.030-SCALE MODEL (45-0) OFTHE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B (MODIFIED) INTHE NASA AMES RESEARCH CENTER UNITARY PLAN WIND TUNNEL ( )
2245	147,619	V-02	OA161A/B/C	RESULTS OF AN INVESTIGATION TO DETERMINE LOCAL FLOW CHARACTERISTICS AT THE AIR DATA PROBE LOCATIONS USING AN 0.030-SCALE MODEL (45-0) OFTHE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B (MODIFIED) INTHE NASA AMES RESEARCH CENTER UNITARY PLAN WIND TUNNEL ( )
2246	144,600		LA65	LOW SUBSONIC AERODYNAMIC CHARACTERISTICS OF FIVE IRREGULAR PLANFORM WINGS WITH SYSTEMATICALLY VARYING WING FILLET GEOMETRY TESTED IN THENASA/AMES 12-FOOT PRESSURE TUNNEL (LA65)
2247	141,834		OA160	RESULTS OF AN INVESTIGATION OF HYPERSONIC VISCOUS INTERACTION EFFECTS OF THE SPACE SHUTTLE ORBITER USING A 0.01/ SCALE MODEL (51-0) IN THE AEDC-VKF TUNNEL F
2248	144,599		IH48	RESULTS OF HEAT TRANSFER TESTS OF A 0.0175-SCALE SPACE SHUTTLE VEHICLE 5 MODEL (60-OTS) IN THE NASA-AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL (TEST IH48)
2249	151,775		IH33	RESULTS OF SPACE SHUTTLE HEAT TRANSFER TESTS USING A 0.01-SCALE MODEL (37-OT) IN THE CALSPAN HYPERSONIC SHOCK TUNNEL (TEST IH33)
2250	141,539		OH43	RESULTS OF CONVECTIVE HEATING TESTS OF A LONGITUDINAL GAP ON THE ROCKWELL FLAT PLATE MODEL (15-0, INSERT VII) IN THE NASA/AMES 3.5 FOOT HYPERSONIC WIND TUNNEL (TEST OH43)

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2251	141,540		OH9	PESULTS OF TESTS ON A ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER (-139 CONFIGURATION) 0.0175-SCALE MODEL (NO. 29-0) IN AEDC TUNNEL B TO DETERMINE BOUNDARY LAYER CHARACTERISTICS
2252	141,546		OH25A	HEAT TRANSFER PHASE CHANGE PAINT TESTS OF 0.0175-SCALE MODELS (NOS. 21-0 AND 46-0) OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER IN THE AEDC TUNNEL B HYPERSONIC WIND TUNNEL (TEST OH25A)
2253	144,833		IA125	AN INVESTIGATION IN THE MSFC TNT TO DETERMINE SPOILER EFFECTS ON WING LOADS AND ELEVON HINGE MOMENTS UTILIZING 0.004-SCALE MODELS (77-0 AND 74-OTS) OF THE SHUTTLE VEHICLE 5 CONFIGURATION
2254	144,619	V-01	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	144,620	V-02	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	144,621	V-03	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	144,622	V-04	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	144,623	V-05	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	144,624	V-06	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	144,625	V-07	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)

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2254	144,626	V-08	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	144,627	V-09	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	144,628	V-10	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	147,601	V-11	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	147,602	V-12	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2254	147,603	V-13	OA148/OA148P	TERMINAL AREA ENERGY MANAGEMENT REGIME INVESTIGATIONS UTILIZING AN 0.030-SCALE MODEL (47-0) OF THE SPACE SHUTTLE VEHICLE ORBITER CONFIGURATION 140A/B/C/R IN THE AMES RESEARCH CENTER 11 X 11 FOOT TRANSONIC WIND TUNNEL (OA148)
2255				SHADOWGRAPHS OF AIR FLOW OVER PROSPECTIVE SPACE SHUTTLE CONFIGURATIONS AT MACH NUMBERS FROM 0.8 TO 1.4
2256			LA68	** DOCUMENTATION NOT COMPLETE **
2257	151,369		LA69	RESULTS OF A DRAG REDUCTION INVESTIGATION ON AN 0.010-SCALE MODEL OFTHE SPACE SHUTTLE VEHICLE (72-OTS) LAUNCH CONFIGURATION TESTED IN THE LARC 8-FOOT TRANSONIC PRESSURE TUNNEL FOR THE MACH RANGE OF 0.35 TO 1.20
2258	151,045	V-01	IA72	INVESTIGATIONS ON A 0.020-SCALE JET PLUME MODEL (88-OTS) OF THE ROCKWELL INTERNATIONAL INTEGRATED SSV CONFIGURATION 14DC (MODIFIED) IN THE 11-FOOT TRANSONIC WIND TUNNEL
2258	151,046	V-02	IA72	INVESTIGATIONS ON A 0.020-SCALE JET PLUME MODEL (88-OTS) OF THE ROCKWELL INTERNATIONAL INTEGRATED SSV CONFIGURATION 14DC (MODIFIED) IN THE 11-FOOT TRANSONIC WIND TUNNEL

CHRYSLER DATA MANAGEMENT SERVICES(DMS)
SPACE SHUTTLE WIND TUNNEL TEST PROGRAM
DATA REPORT DOCUMENTATION

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2258	151,047	V-03	IA72	INVESTIGATIONS ON A 0.020-SCALE JET PLUME MODEL (88-OTS) OF THE ROCKWELL INTERNATIONAL INTEGRATED SSV CONFIGURATION 14DC (MODIFIED) IN THE 11-FOOT TRANSONIC WIND TUNNEL
2258	151,048	V-04	IA72	INVESTIGATIONS ON A 0.020-SCALE JET PLUME MODEL (88-OTS) OF THE ROCKWELL INTERNATIONAL INTEGRATED SSV CONFIGURATION 14DC (MODIFIED) IN THE 11-FOOT TRANSONIC WIND TUNNEL
2258	151,049	V-05	IA72	INVESTIGATIONS ON A 0.020-SCALE JET PLUME MODEL (88-OTS) OF THE ROCKWELL INTERNATIONAL INTEGRATED SSV CONFIGURATION 14DC (MODIFIED) IN THE 11-FOOT TRANSONIC WIND TUNNEL
2258	151,050	V-06	IA72	INVESTIGATIONS ON A 0.020-SCALE JET PLUME MODEL (88-OTS) OF THE ROCKWELL INTERNATIONAL INTEGRATED SSV CONFIGURATION 14DC (MODIFIED) IN THE 11-FOOT TRANSONIC WIND TUNNEL
2258	151,051	V-07	IA72	INVESTIGATIONS ON A 0.020-SCALE JET PLUME MODEL (88-OTS) OF THE ROCKWELL INTERNATIONAL INTEGRATED SSV CONFIGURATION 14DC (MODIFIED) IN THE 11-FOOT TRANSONIC WIND TUNNEL
2258	151,052	V-08	IA72	INVESTIGATIONS ON A 0.020-SCALE JET PLUME MODEL (88-OTS) OF THE ROCKWELL INTERNATIONAL INTEGRATED SSV CONFIGURATION 14DC (MODIFIED) IN THE 11-FOOT TRANSONIC WIND TUNNEL
2258	151,053	V-09	IA72	INVESTIGATIONS ON A 0.020-SCALE JET PLUME MODEL (88-OTS) OF THE ROCKWELL INTERNATIONAL INTEGRATED SSV CONFIGURATION 14DC (MODIFIED) IN THE 11-FOOT TRANSONIC WIND TUNNEL
2259	TASK CANCELLED		LA60A	
2260			LA60B/LA60C	** DOCUMENTATION NOT COMPLETE **
2261	167,364	V-01	OA100	RESULTS OF TESTS USING A 0.36-SCALE MODEL(76-O) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE NASA/AMES RESEARCH CENTER 40 BY 80-FOOT SUBSONIC WIND TUNNEL (OA100)
2261	167,365	V-02	OA100	RESULTS OF TESTS USING A 0.36-SCALE MODEL(76-O) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE NASA/AMES RESEARCH CENTER 40 BY 80-FOOT SUBSONIC WIND TUNNEL (OA100)
2262	147,630	V-01	CA6	RESULTS OF A CARRIER AIRCRAFT VERIFICATION TEST IN THE BOEING 8 X 1 2 FOOT TRANSONIC TUNNEL USING A 0.03-SCALE 747 CAM/ORBITER MODEL 45-O

CHRYSLER DATA MANAGEMENT SERVICES(DMS)
SPACE SHUTTLE WIND TUNNEL TEST PROGRAM
DATA REPORT DOCUMENTATION

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2262	147,631	V-02	CA6	RESULTS OF A CARRIER AIRCRAFT VERIFICATION TEST IN THE BOEING 8 X 1 2 FOOT TRANSONIC TUNNEL USING A 0.03-SCALE 747 CAM/ORBITER MODEL 45-0
2263	144,596		OH74	RESULTS OF HEAT TRANSFER TESTS ON A 0.0175-SCALE SPACE SHUTTLE ORBITER MODEL (56-0) IN THE AEDC VKF 'B' HYPERSONIC WIND TUNNEL (OH74)
2264	141,843		LA62	TRANSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE (REMOTELY CONTROLLED ELEVON) MODEL 49-0 OF THE SPACE SHUTTLE ORBITER TESTED IN THE NASA/LARC 8-FOOT TPT (LA62)
2265	141,832		OA159	RESULTS OF TESTS USING A 0.030-SCALE MODEL (45-0) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE NASA/ARC 12-FOOT PRESSURE TUNNEL (OA159)
2266	144,607		LA67	TRANSONIC-SUPERSONIC HIGH REYNOLDS NUMBER STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE (REMOTELY CONTROLLED ELEVON) MODEL 44-0 OF THE SPACE SHUTTLE ORBITER TESTED IN THE VSD HIGH SPEED WIND TUNNEL
2267	147,604	V-01	MA22	RESULTS OF TEST MA22 IN THE NASA/LARC 31-INCH CFHT ON AN 0.010-SCALEMODEL (32-0) OF THE SPACE SHUTTLE CONFIGURATION 3 TO DETERMINE RCS JET FLOW FIELD INTERACTION AND TO INVESTIGATE RT REAL GAS EFFECTS
2267	147,605	V-02	MA22	RESULTS OF TEST MA22 IN THE NASA/LARC 31-INCH CFHT ON AN 0.010-SCALEMODEL (32-0) OF THE SPACE SHUTTLE CONFIGURATION 3 TO DETERMINE RCS JET FLOW FIELD INTERACTION AND TO INVESTIGATE RT REAL GAS EFFECTS
2267	147,606	V-03	MA22	RESULTS OF TEST MA22 IN THE NASA/LARC 31-INCH CFHT ON AN 0.010-SCALEMODEL (32-0) OF THE SPACE SHUTTLE CONFIGURATION 3 TO DETERMINE RCS JET FLOW FIELD INTERACTION AND TO INVESTIGATE RT REAL GAS EFFECTS
2267	147,607	V-04	MA22	RESULTS OF TEST MA22 IN THE NASA/LARC 31-INCH CFHT ON AN 0.010-SCALEMODEL (32-0) OF THE SPACE SHUTTLE CONFIGURATION 3 TO DETERMINE RCS JET FLOW FIELD INTERACTION AND TO INVESTIGATE RT REAL GAS EFFECTS
2268	151,396	V-01	CA9/CA9P	RESULTS OF AN INVESTIGATION OF AERODYNAMIC FORCES, MOMENTS, AND PRESSURES ON 0.03-SCALE MODELS OF THE MATED SPACE SHUTTLE ORBITER AND CARRIER AIRCRAFT (MODEL NUMBERS AX1319P-1 AND 47-0) IN THE BOEING TRANSONIC WIND TUNNEL (CA9)
2268	151,397	V-02	CA9/CA9P	RESULTS OF AN INVESTIGATION OF AERODYNAMIC FORCES, MOMENTS, AND PRESSURES ON 0.03-SCALE MODELS OF THE MATED SPACE SHUTTLE ORBITER AND CARRIER AIRCRAFT (MODEL NUMBERS AX1319P-1 AND 47-0) IN THE BOEING TRANSONIC WIND TUNNEL (CA9)

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2268	151,398	V-03	CA9/CA9P	RESULTS OF AN INVESTIGATION OF AERODYNAMIC FORCES, MOMENTS, AND PRESSURES ON 0.03-SCALE MODELS OF THE MATED SPACE SHUTTLE ORBITER AND CARRIER AIRCRAFT (MODEL NUMBERS AX1319P-1 AND 47-0) IN THE BOEING TRANSONIC WIND TUNNEL (CA9)
2268	151,399	V-04	CA9/CA9P	RESULTS OF AN INVESTIGATION OF AERODYNAMIC FORCES, MOMENTS, AND PRESSURES ON 0.03-SCALE OODELS OF THE MATED SPACE SHUTTLE ORBITER AND CARRIER AIRCRAFT (MODEL NUMBERS AX1319P-1 AND 47-0) IN THE BOEING TRANSONIC WIND TUNNEL (CA9)
2268	151,400	V-05	CA9/CA9P	RESULTS OF AN INVESTIGATION OF AERODYNAMIC FORCES, MOMENTS, AND PRES URES ON 0.03-SCALE MODELS OF THE MATED SPACE SHUTTLE ORBITER AND CARRIER AIRCRAFT (MODEL NUMBERS AX1319P-1 AND 47-0) IN THE BOEING TRANSONIC WIND TUNNEL (CA9)
2269	147,624		LA70	TRANSONIC HIGH REYNOLDS NUMBER STABILITY AND CONTROL CHARACTERISTICSOF A 0.015-SCALE REMOTELY CONTROLLED ELEVON MODEL (44-0) OF THE SPACE SHUTTLE ORBITER TESTED IN THE CALSPAN 8-FOOT TWT
2270	144,579		LA63A	LOW SUPERSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE REMOTELY CONTROLLED ELEVON MODEL (49-0) OF THE SPACE SHUTTLE ORBITER (LA63A)
2271	151,044		LA71A/B	SUPERSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015 SCALE MODEL 69-0 OF THE SPACE SHUTTLE ORBITER WITH FOREBODY RSI MODIFICATIONS IN THE NASA/LARC 4-FOOT UPWT (LEGS 1 AN) 2)
2272	151,077	V-01	IA114	RESULTS OF AN INVESTIGATION OF EXTERNAL TANK SEPARATION EFFECTS USING AN 0.010-SCALE MODEL (52-OT) SPACE SHUTTLE VEHICLE IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER VON KARMAN FACILITY TUNNEL B
2272	151,078	V-02	IA114	RESULTS OF AN INVESTIGATION OF EXTERNAL TANK SEPARATION EFFECTS USING AN 0.010-SCALE MODEL (52-OT) SPACE SHUTTLE VEHICLE IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER VON KARMAN FACILITY TUNNEL B
2273	144,612	V-01	CA26	RESULTS OF AN AERODYNAMIC INVESTIGATION OF A SPACE SHUTTLE ORBITER/747 CARRIER FLIGHT TEST CONFIGURATION TO DETERMINE SEPARATION CHARACTERISTICS UTILIZING 0.0125-SCALE MODELS (48-0/AX1318I-1) IN THE LTV 4X4-FOOT HIGH SPEED WIND TUNNEL (CA26)
2273	144,613	V-02	CA26	RESULTS OF AN AERODYNAMIC INVESTIGATION OF A SPACE SHUTTLE ORBITER/747 CARRIER FLIGHT TEST CONFIGURATION TO DETERMINE SEPARATION CHARACTERISTICS UTILIZING 0.0125-SCALE MODELS (48-0/AX1318I-1) IN THE LTV 4X4-FOOT HIGH SPEED WIND TUNNEL (CA26)

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2273	144,614	V-03	CA26	RESULTS OF AN AERODYNAMIC INVESTIGATION OF A SPACE SHUTTLE ORBITER/747 CARRIER FLIGHT TEST CONFIGURATION TO DETERMINE SEPARATION CHARACTERISTICS UTILIZING 0.0125-SCALE MODELS (48-O/AX1318I-1) IN THE LTV 4X4-FOOT HIGH SPEED WIND TUNNEL (CA26)
2273	144,615	V-04	CA26	RESULTS OF AN AERODYNAMIC INVESTIGATION OF A SPACE SHUTTLE ORBITER/747 CARRIER FLIGHT TEST CONFIGURATION TO DETERMINE SEPARATION CHARACTERISTICS UTILIZING 0.0125-SCALE MODELS (48-O/AX1318I-1) IN THE LTV 4X4-FOOT HIGH SPEED WIND TUNNEL (CA26)
2273	144,616	V-05	CA26	RESULTS OF AN AERODYNAMIC INVESTIGATION OF A SPACE SHUTTLE ORBITER/747 CARRIER FLIGHT TEST CONFIGURATION TO DETERMINE SEPARATION CHARACTERISTICS UTILIZING 0.0125-SCALE MODELS (48-O/AX1318I-1) IN THE LTV 4X4-FOOT HIGH SPEED WIND TUNNEL (CA26)
2274	144,593		FA14	AN INVESTIGATION OF DRAG REDUCTION FAIRINGS ON THE SPACE SHUTTLE VEHICLE 5 CONFIGURATION (MODEL 74-OTS) IN THE MSFC 14-INCH TRISONIC WIND TUNNEL
2275	144,603	V-01	CA23B	RESULTS OF AN EXPERIMENTAL INVESTIGATION TO DETERMINE SEPARATION CHARACTERISTICS FOR THE ORBITER/747 USING A 0.0125-SCALE MODEL (48-O AX1318I-1 747) IN THE AMES RESEARCH CENTER 14-FOOT WIND TUNNEL (CA23B)
2275	144,604	V-02	CA23B	RESULTS OF AN EXPERIMENTAL INVESTIGATION TO DETERMINE SEPARATION CHARACTERISTICS FOR THE ORBITER/747 USING A 0.0125-SCALE MODEL (48-O AX1318I-1 747) IN THE AMES RESEARCH CENTER 14-FOOT WIND TUNNEL (CA23B)
2276	151,055		FH13	HEAT TRANSFER AND SURFACE PRESSURE DATA OBTAINED ON A .0429 SCALE MODEL SSV EXTERNAL TANK NOSE SECTION AT MACH NUMBERS FROM 2.5 TO 5.5 (FH13)
2277	144,579		SA13F	FORCE TEST OF A 0.88 PERCENT SCALE 142-INCH DIAMETER SOLID ROCKET BOOSTER (MSFC MODEL NUMBER 461) IN THE NASA/MSFC HIGH REYNOLDS NUMBER WIND TUNNEL
2278	TASK CANCELLED		LA61	LOW-SUBSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.010-SCALE REMOTELY CONTROLLED ELEVON MODEL (49-O) OF THE SPACE SHUTTLE ORBITER IN THE LANGLEY RESEARCH CENTER LOW TURBULENCE PRESSURE TUNNEL
2279	144,606		LA63B	HIGH SUPERSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE (REMOTELY CONTROLLED ELEVON) MODEL 49-O OF THE SPACE SHUTTLE ORBITER TESTED IN THE NASA/LARC 4-FOOT UPWT(LEG 2)
2280	144,582		LA28	HEAT-FLUX GAGE MEASUREMENTS ON A FLAT PLATE AT A MACH NUMBER OF 4.6 IN THE VSD HIGH SPEED WIND TUNNEL--A FEASIBILITY TEST (LA28)

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2281	147,621		LA66	SUBSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE (REMOTELY CONTROLLED ELEVON) MODEL 44-0 OF THE SPACE SHUTTLE ORBITER TESTED IN THE NASA/ARC 12-FOOT PRESSURE TUNNEL (LA66)
2282	151,407		IH34	BASE PRESSURE AND HEAT TRANSFER TESTS OF THE 0.0225-SCALE SPACE SHUTTLE PLUME SIMULATION MODEL 19-OTS IN THE NASA-LEWIS 10X10 FOOT SWT
2283	147,649		MA14	A LOW SPEED WIND TUNNEL TEST OF A 0.050 SCALE MODEL OF SHUTTLE ORBITER (MODEL 089B) TO INVESTIGATE THE LONGITUDINAL AND LATERAL DIRECTIONAL EFFECTS OF CANARD AND TAIL CONFIGURATIONAL MODIFICATIONS IN THE LTV LSWT
2284	151,035	V-01	IS2A/B	AERODYNAMIC NOISE OF THE 0.035-SCALE INTEGRATED SPACE SHUTTLE VEHICLE MODEL (84-OTS) IN THE NASA-AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS (IS2A/B)
2284	151,036	V-02	IS2A/B	AERODYNAMIC NOISE OF THE 0.035-SCALE INTEGRATED SPACE SHUTTLE VEHICLE MODEL (84-OTS) IN THE NASA-AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS (IS2A/B)
2285	144,595		OH50A	RESULTS OF TESTS USING THE PHASE CHANGE PAINT TECHNIQUE ON 0.04 SCALE 50 PERCENT FOREBODY MODELS (82-0) OF THE ROCKWELL SPACE SHUTTLE ORBITER
2286	147,625		OA220	RESULTS OF AN AIR PROBE INVESTIGATION UTILIZING A 0.10 SCALE ORBITER (MODEL 57-0) FOREBODY IN THE AMES RESEARCH CENTER 14 FOOT WIND TUNNEL (OA220)
2287			OS13	** DOCUMENTATION NOT COMPLETE **
2288	151,384		OH64	RESULTS OF BASE HEATING INVESTIGATIONS ON A 0.04 SCALE SPACE SHUTTLEORBITER BASE (MODEL 25-0) IN THE NASA/LARC SPACE POWER FACILITY
2289	147,611	V-01	OA163	RESULTS OF A LANDING LOADS TEST USING A 0.0405-SCALE MODEL (16-0) OF THE SPACE SHUTTLE ORBITER IN THE ROCKWELL INTERNATIONAL NAAL WIND TUNNEL (OA163)
2289	147,612	V-02	OA163	RESULTS OF A LANDING LOADS TEST USING A 0.0405-SCALE MODEL (16-0) OF THE SPACE SHUTTLE ORBITER IN THE ROCKWELL INTERNATIONAL NAAL WIND TUNNEL (OA163)
2289	147,613	V-03	OA163	RESULTS OF A LANDING LOADS TEST USING A 0.0405-SCALE MODEL (16-0) OF THE SPACE SHUTTLE ORBITER IN THE ROCKWELL INTERNATIONAL NAAL WIND TUNNEL (OA163)
2289	147,614	V-04	OA163	RESULTS OF A LANDING LOADS TEST USING A 0.0405-SCALE MODEL (16-0) OF THE SPACE SHUTTLE ORBITER IN THE ROCKWELL INTERNATIONAL NAAL WIND TUNNEL (OA163)

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2290	147,641	V-01	CA8	MATED AERODYNAMIC CHARACTERISTICS INVESTIGATION FOR THE 0.04 SCALE 747 CAM AND THE 0.0405 SCALE SPACE SHUTTLE ORBITER IN THE NASA LANGLEY V/STOL TRANSITION RESEARCH WIND TUNNEL
2290	147,642	V-02	CA8	MATED AERODYNAMIC CHARACTERISTICS INVESTIGATION FOR THE 0.04 SCALE 747 CAM AND THE 0.0405 SCALE SPACE SHUTTLE ORBITER IN THE NASA LANGLEY V/STOL TRANSITION RESEARCH WIND TUNNEL
2290	147,643	V-03	CA8	MATED AERODYNAMIC CHARACTERISTICS INVESTIGATION FOR THE 0.04 SCALE 747 CAM AND THE 0.0405 SCALE SPACE SHUTTLE ORBITER IN THE NASA LANGLEY V/STOL TRANSITION RESEARCH WIND TUNNEL
2291			LA79	** DOCUMENTATION NOT COMPLETE **
2292	TM-X72661		LA36B	** TO BE PUBLISHED AT LARC **
2293	151,381		IA40	RESULTS OF TESTS USING A 0.010-SCALE SSV MODEL 75-OTS IN THE AEDC VKF TUNNEL A
2294	160,822	V-01	OA172	RESULTS OF TESTS OF A SPACE SHUTTLE ORBITER FERRY CONFIGURATION USING A 140A/B 0.0405-SCALE MODEL (43-O) IN THE ROCKWELL INTERNATIONAL 7.75 X 11 FOOT LOW SPEED WIND TUNNEL (OA172)
2294	160,823	V-02	OA172	RESULTS OF TESTS OF A SPACE SHUTTLE ORBITER FERRY CONFIGURATION USING A 140A/B 0.0405-SCALE MODEL (43-O) IN THE ROCKWELL INTERNATIONAL 7.75 X 11 FOOT LOW SPEED WIND TUNNEL (OA172)
2295	151,069	V-01	IH41B	RESULTS OF AN INVESTIGATION OF THE SPACE SHUTTLE INTEGRATED VEHICLE AERODYNAMIC HEATING CHARACTERISTICS OBTAINED USING THE 0.0175-SCALE MODEL 60-OTS IN AEDC TUNNEL A DURING TESTS IH41B
2295	151,070	V-02	IH41B	RESULTS OF AN INVESTIGATION OF THE SPACE SHUTTLE INTEGRATED VEHICLE AERODYNAMIC HEATING CHARACTERISTICS OBTAINED USING THE 0.0175-SCALE MODEL 60-OTS IN AEDC TUNNEL A DURING TESTS IH41B
2295	151,071	V-03	IH41B	RESULTS OF AN INVESTIGATION OF THE SPACE SHUTTLE INTEGRATED VEHICLE AERODYNAMIC HEATING CHARACTERISTICS OBTAINED USING THE 0.0175-SCALE MODEL 60-OTS IN AEDC TUNNEL A DURING TESTS IH41B
2295	151,072	V-04	IH41B	RESULTS OF AN INVESTIGATION OF THE SPACE SHUTTLE INTEGRATED VEHICLE AERODYNAMIC HEATING CHARACTERISTICS OBTAINED USING THE 0.0175-SCALE MODEL 60-OTS IN AEDC TUNNEL A DURING TESTS IH41B

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2295	151,073	V-05	IH41B	RESULTS OF AN INVESTIGATION OF THE SPACE SHUTTLE INTEGRATED VEHICLE AERODYNAMIC HEATING CHARACTERISTICS OBTAINED USING THE 0.0175-SCALE MODEL 60-OTS IN AEDC TUNNEL A DURING TESTS IH41B
2296	147,609	V-01	LA81	SHUTTLE MODEL TAILCONE PRESSURE DISTRIBUTION AT LOW SUBSONIC SPEEDS OF A 0.03614-SCALE MODEL IN THE NASA/LARC LOW TURBULENCE PRESSURE TUNNEL (LA81)
2296	147,610	V-02	LA81	SHUTTLE MODEL TAILCONE PRESSURE DISTRIBUTION AT LOW SUBSONIC SPEEDS OF A 0.03614-SCALE MODEL IN THE NASA/LARC LOW TURBULENCE PRESSURE TUNNEL (LA81)
2297	147,628		LA45A/B	HIGH SUPERSONIC AERODYNAMIC CHARACTERISTICS OF FIVE IRREGULAR PLANFORM WINGS WITH SYSTEMATICALLY VARYING WING FILLET GEOMETRY TESTED IN THE NASA/LARC 4-FOOT UPWT (LEG 2) (LA45A/B)
2298	151,409		LA73A/LA73B	LOW SPEED STABILITY AND CONTROL CHARACTERISTICS OF A 0.015 SCALE MODEL 69-0 OF THE SPACE SHUTTLE ORBITER WITH FOREBODY RSI MODIFICATIONSIN THE NASA/LARC LOW TURBULENCE PRESSURE TUNNEL (LA73A/B)
2299	TM-X3497		LA80	DYNAMIC STABILITY CHARACTERISTICS OF THE COMBINATION SPACE SHUTTLE OORBITER AND FERRY COMBINATION
2300	147,629		LA61B	LOW-SUBSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE REMOTELY CONTROLLED ELEVON MODEL (44-0) OF THE SPACE SHUTTLE ORBITER IN THE LANGLEY RESEARCH CENTER LOW TURBULENCE PRESSURE TUNNEL
2301	144,605		OH54A	RESULTS OF PHASE CHANGE PAINT HEAT TRANSFER TESTS UTILIZING 0.040 SCALE 50 PERCENT FOREBODY MODELS (NO. 82-0) OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER IN AEDC VKF HYPERSONIC TUNNEL B
2302	167,340	V-01	OA174	RESULTS OF TESTS USING A 0.36-SCALE MODEL(76-0) OF THE SPACE SHUTTLE ORBITER VEHICLE 101 IN THE NASA/AMES RESEARCH CENTER'S 40 X 80 SUBSONIC WIND TUNNEL (OA174)
2302	167,341	V-02	OA174	RESULTS OF TESTS USING A 0.36-SCALE MODEL(76-0) OF THE SPACE SHUTTLE ORBITER VEHICLE 101 IN THE NASA/AMES RESEARCH CENTER'S 40 X 80 SUBSONIC WIND TUNNEL (OA174)
2303	144,618		OH75	RESULTS OF PHASE CHANGE PAINT TESTS OF 0.040 SCALE 50 PERCENT FOREBODY MODELS (82-0) OF THE SPACE SHUTTLE ORBITER IN THE AEDC VKF 'B' HYPERSONIC WIND TUNNEL

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2304	150,846		OA173	RESULTS OF TESTS TO EVALUATE ARC 40X80-FOOT TUNNEL SUPPORT STRUT TARES ON THE SPACE SHUTTLE VEHICLE WITH TAIL CONE USING A 0.03-SCALE MODEL (45-0) IN THE NASA/ARC 12-FOOT PRESSURE WIND TUNNEL (OA173)
2305	151,059	V-01	LA76	HIGH REYNOLDS NUMBER TRANSONIC STABILITY AND CONTROL CHARACTERISTICSOF A 0.015 SCALE(REMOTELY CONTROLLED ELEVON) MODEL 44-0 OF THE SPACESHUTTLE ORBITER TESTED IN THE VSD HICH SPEED TUNNEL(LA76)
2305	151,060	V-02	LA76	HIGH REYNOLDS NUMBER TRANSONIC STABILITY AND CONTROL CHARACTERISTICSOF A 0.015 SCALE(REMOTELY CONTROLLED ELEVON) MODEL 44-0 OF THE SPACESHUTTLE ORBITER TESTED IN THE VSD HIGH SPEED TUNNEL(LA76)
2306	167,354	V-01	IA135A/B/C	RESULTS OF TESTS ON THE SPACE SHUTTLE LAUNCH CONFIGURATION USING THE 0.03 SCALE MODEL 47-OTS IN THE NASA/AMES UNITARY PLAN WIND TUNNEL (IA135A/B/C)
2306	167,355	V-02	IA135A/B/C	RESULTS OF TESTS ON THE SPACE SHUTTLE LAUNCH CONFIGURATION USING THE 0.03 SCALE MODEL 47-OTS IN THE NASA/AMES UNITARY PLAN WIND TUNNEL (IA135A/B/C)
2306	167,356	V-03	IA135A/B/C	RESULTS OF TESTS ON THE SPACE SHUTTLE LAUNCH CONFIGURATION USING THE 0.03 SCALE MODEL 47-OTS IN THE NASA/AMES UNITARY PLAN WIND TUNNEL (IA135A/B/C)
2307	160,840	V-01	CA14A	RESULTS OF EXPERIMENTAL AERODYNAMIC INVESTIGATION ON A 0.03 SCALE MODEL BOEING 747 CAM WITH SPACE SHUTTLE ORBITER IN THE BOEING 8X12 FOOT TRANSONIC WIND TUNNEL (CA14A)
2307	160,841	V-02	CA14A	RESULTS OF EXPERIMENTAL AERODYNAMIC INVESTIGATION ON A 0.03 SCALE MODEL BOEING 747 CAM WITH SPACE SHUTTLE ORBITER IN THE BOEING 8X12 FOOT TRANSONIC WIND TUNNEL (CA14A)
2308	147,636		IH5	AN EXPERIMENTAL DETERMINATION IN THE CALSPAN LUDWIEG TUBE OF THE BASE ENVIRONMENT OF THE INTEGRATED SPACE SHUTTLE VEHICLE AT SIMULATED MACH 4.5 FLIGHT CONDITIONS (TEST IH5 OF MODEL 19-OTS)
2309	147,644		LA72	TRANSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015 SCALE MODEL 69-0 OF THE SPACE SHUTTLE ORBITER WITH FOREBODY RSI MODIFICATION IN THE NASA/LARC 8-FOOT TPT (LA72)
2310	151,083	V-01	SA14FB	REENTRY STATIC STABILITY CHARACTERISTICS OF A 0.00548 SCALE MODEL OFA RIGHT HAND 146-INCH DIAMETER SOLID ROCKET BOOSTER (MSFC MODEL 486)REENTRY CONFIGURATION AS DETERMINED FROM TESTS IN THE NASA/MSFC 14-INCH TRISONIC WIND TUNNEL

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2310	151,084	V-02	SA14FB	REENTRY STATIC STABILITY CHARACTERISTICS OF A 0.00548 SCALE MODEL OFA RIGHT HAND 146-INCH DIAMETER SOLID ROCKET BOOSTER (MSFC MODEL 486)REENTRY CONFIGURATION AS DETERMINED FROM TESTS IN THE NASA/MSFC 14-INCH TRISONIC WIND TUNNEL
2311	147,620		LA78/LA87/LA88	RESULTS FROM INVESTIGATIONS IN THREE NASA/LARC HYPERSONIC WIND TUNNELS ON A 0.004-SCALE MODEL SPACE SHUTTLE ORBITER (MODEL 13P-O)TO DET ERMINE REAL GAS EFFECTS (LA78, LA87, LA88)
2312	151,075	V-01	IH47	RESULTS OF AN INVESTIGATION OF THE SPACE SHUTTLE SOLID ROCKET BOOSTER AERODYNAMIC HEATING CHARACTERISTICS OBTAINED USING THE 0.0175-SCALE MODEL 60-OTS IN AEDC TUNNEL A DURING TESTS IH47
2312	151,076	V-02	IH47	RESULTS OF AN INVESTIGATION OF THE SPACE SHUTTLE SOLID ROCKET BOOSTER AERODYNAMIC HEATING CHARACTERISTICS OBTAINED USING THE 0.0175-SCALE MODEL 60-OTS IN AEDC TUNNEL A DURING TESTS IH47
2313	151,041	V-01	FH14	RESULTS OF WIND TUNNEL TESTS TO DETERMINE HEAT TRANSFER RATES ON A .0275 SCALE SPACE SHUTTLE EXTERNAL TANK WITH A 10 DEG/40 DEG DOUBLE CONE-OGIVE NOSE IN THE NASA/ARC 3.5 HYPERSONIC TUNNEL
2313	151,042	V-02	FH14	RESULTS OF WIND TUNNEL TESTS TO DETERMINE HEAT TRANSFER RATES ON A .0275 SCALE SPACE SHUTTLE EXTERNAL TANK WITH A 10 DEG/40 DEG DOUBLE CONE-OGIVE NOSE IN THE NASA/ARC 3.5 HYPERSONIC TUNNEL
2313	151,043	V-03	FH14	RESULTS OF WIND TUNNEL TESTS TO DETERMINE HEAT TRANSFER RATES ON A .0275 SCALE SPACE SHUTTLE EXTERNAL TANK WITH A 10 DEG/40 DEG DOUBLE CONE-OGIVE NOSE IN THE NASA/ARC 3.5 HYPERSONIC TUNNEL
2314	151,406		OA176	INVESTIGATION OF SUPPORT SYSTEM EFFECTS ON ORBITER LOW SPEED AEORDYNAMIC CHARACTERISTICS USING 0.0405-SCALE MODEL 43-O IN THE NAAL LOW SPEED WIND TUNNEL
2315	147,623		IA141	RESULTS OF AN INVESTIGATION OF REYNOLDS NUMBER EFFECTS ON INTEGRATEDVEHICLE ELEVON HINGE MOMENTS AND WING PANEL LOADS OBTAINED WITH 0.010-SCALE MODEL 72-OTS IN THE ROCKWELL TRISONIC WIND TUNNEL
2316	147,622		IA137	RESULTS OF TEST IA137 IN THE NASA/ARC 14 FOOT TRANSONIC WIND TUNNEL OF THE 0.07 SCALE EXTERNAL TANK FOREBODY (MODEL 68-T) TO DETERMINE AUXILIARY AERODYNAMIC DATA SYSTEM FEASIBILITY

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2317	151,787		OH53A	RESULTS OF TESTS TO DETERMINE REACTION CONTROL SYSTEM (RCS) NOZZLE EFFECTS ON THE ORBITER FOREBODY ASCENT AERODYNAMIC HEATING RATES USING A 0.04-SCALE MODEL (83-0) IN THE AMES RESEARCH CENTER 3.5 FOOT HYPERSONIC WIND TUNNEL (OH53A)
2318	147,646	V-01	LA75	HIGH SUPERSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE (REMOTELY CONTROLLED ELEVON) MODEL 44-0 SPACE SHUTTLE ORBITER TESTED IN THE NASA/LARC 4-FOOT UPWT (LEG 2) (LA75)
2318	147,647	V-02	LA75	HIGH SUPERSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE (REMOTELY CONTROLLED ELEVON) MODEL 44-0 SPACE SHUTTLE ORBITER TESTED IN THE NASA/LARC 4-FOOT UPWT (LEG 2) (LA75)
2319	151,771		IH43	HEAT TRANSFER AND PRESSURE TESTS ON A 0.01-SCALE SPACE SHUTTLE MODEL (59-OT) IN THE CALSPAN HYPERVELOCITY SHOCK TUNNELS (IH43)
2320	151,390	V-01	OA169	RESULTS OF TESTS USING A 0.0125-SCALE MODEL(70-OT)OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE AEDC VKF TUNNEL B (OA169)
2320	151,391	V-02	OA169	RESULTS OF TESTS USING A 0.0125-SCALE MODEL(70-OT)OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE AEDC VKF TUNNEL B (OA169)
2320	151,392	V-03	OA169	RESULTS OF TESTS USING A 0.0125-SCALE MODEL(70-OT)OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE AEDC VKF TUNNEL B (OA169)
2321	151,410	V-01	OH69	RESULTS OF TEST OH69 OBTAINED IN THE AEDC VKF HYPERSONIC TUNNEL B USING THE INFRARED SCANNING METHOD TO OBTAIN HEAT TRANSFER DATA ON THE 0.040 SCALE MODEL 82-0 OF THE SPACE SHUTTLE FOREBODY
2321	151,411	V-02	OH69	RESULTS OF TEST OH69 OBTAINED IN THE AEDC VKF HYPERSONIC TUNNEL B USING THE INFRARED SCANNING METHOD TO OBTAIN HEAT TRANSFER DATA ON THE 0.040 SCALE MODEL 82-0 OF THE SPACE SHUTTLE FOREBODY
2322	160,847		OA228	RESULTS OF TEST OA228 USING THE SSV VEHICLE 102 0.10 SCALE FOREBODY MODEL NO. 57-0 IN THE NAAL LOW SPEED WIND TUNNEL
2323	151,039		IA94A	RESULTS OF INVESTIGATIONS CONDUCTED IN THE LARC 4-FOOT UNITARY PLAN WIND TUNNEL LEG NO. 1 USING THE 0.010-SCALE 72-OTS MODEL OF THE SPACE SHUTTLE INTEGRATED VEHICLE
2324	151,040		IA94B	RESULTS OF INVESTIGATIONS CONDUCTED IN THE LARC 4-FOOT UNITARY PLAN WIND TUNNEL LEG NO. 2 USING THE 0.010-SCALE 72-OTS MODEL OF THE SPACE SHUTTLE INTEGRATED VEHICLE

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2325	147,645		SA14FA	AERODYNAMIC CHARACTERISTICS OF A 0.00563 SCALE 142-INCH DIAMETER SOLID ROCKET BOOSTER (MSFC MODEL 449 AND 480) WITH SIDE MOUNTED STINGS IN THE NASA/MSFC 14 INCH TRISONIC WIND TUNNEL
2326	151,037	V-01	IA93	RESULTS OF INVESTICATIONS CONDUCTED IN THE LARC 8-FOOT TRANSONIC PRESSURE TUNNEL USING THE 0.010-SCALE 72-OTS MODEL OF THE SPACE SHUTTLE INTEGRATED VEHICLE
2326	151,038	V-02	IA93	RESULTS OF INVESTICATIONS CONDUCTED IN THE LARC 8-FOOT TRANSONIC PRESSURE TUNNEL USING THE 0.010-SCALE 72-OTS MODEL OF THE SPACE SHUTTLE INTEGRATED VEHICLE
2327	151,079	V-01	IA22	RESULTS OF TESTS USING 0.0125-SCALE MODEL (70-OT) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE AEDC VKF TUNNEL B
2327	151,080	V-02	IA22	RESULTS OF TESTS USING 0.0125-SCALE MODEL (70-OT) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE AEDC VKF TUNNEL B
2327	151,081	V-03	IA22	RESULTS OF TESTS USING 0.0125-SCALE MODEL (70-OT) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE AEDC VKF TUNNEL B
2328	TN D-8233		LA34	EFFECT OF A SURFACE-TO-GAP TEMPERATURE DISCONTINUITY ON THE HEAT TRANSFER TO REUSABLE SURFACE INSULATION TILE GAPS
2329	160,837		OA224	CALIBRATION RESULTS OF THE BASELINE AIR DATA PROBES AT THE LANGLEY 16-FOOT TRANSONIC WIND TUNNEL USING A 0.10 SCALE ORBITER FOREBODY MODEL 102 LINES (OA224)
2330	147,637		OH52	RESULTS OF A FLOW FIELD SURVEY CONDUCTED USING THE 0.0175 SCALE ORBITER MODEL 29-0 IN THE AEDC VKF TUNNEL B DURING TEST OH52
2331	160,838	V-01	SA11F	STATIC STABILITY AND PRESSURE DATA FROM WIND TUNNEL TESTS OF A .028-SCALE (MSFC MODEL 483) SPACE SHUTTLE SRB AT REENTRY ATTITUDES IN THENASA/ARC UNITARY PLAN WIND TUNNELS (SA11F)
2331	160,839	V-02	SA11F	STATIC STABILITY AND PRESSURE DATA FROM WIND TUNNEL TESTS OF A .028-SCALE (MSFC MODEL 483) SPACE SHUTTLE SRB AT REENTRY ATTITUDES IN THENASA/ARC UNITARY PLAN WIND TUNNELS (SA11F)
2332	151,373		CA13	RESULTS OF AERODYNAMIC FORCE AND MOMENT TESTS OF 0.03-SCALE MODELS (AX1319I-3 AND 45-0) OF THE SPACE SHUTTLE ORBITER AND CARRIER IN THE NASA/ARC 14-FOOT TRANSONIC WIND TUNNEL (CA13)

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2333	151,374	V-01	OA175	WIND TUNNEL TEST OA175 OF THE 0.030-SCALE SSV ORBITER MODEL (47-O) IN THE 11 X 11-FOOT LEG OF THE NASA/ARC UNITARY PLAN WIND TUNNEL (OA175)
2333	151,375	V-02	OA175	WIND TUNNEL TEST OA175 OF THE 0.030-SCALE SSV ORBITER MODEL (47-O) IN THE 11 X 11-FOOT LEG OF THE NASA/ARC UNITARY PLAN WIND TUNNEL (OA175)
2333	151,376	V-03	OA175	WIND TUNNEL TEST OA175 OF THE 0.030-SCALE SSV ORBITER MODEL (47-O) IN THE 11 X 11-FOOT LEG OF THE NASA/ARC UNITARY PLAN WIND TUNNEL (OA175)
2334	147,648		SA16F	AN INVESTIGATION OF THE AERODYNAMIC CHARACTERISTICS OF A 0.00548 SCALE MODEL (MODEL NO. 486) OF THE SPACE SHUTTLE 146-INCH DIAMETER SOLID ROCKET BOOSTER AT ANGLES OF ATTACK FROM 113 TO 180 DEGREES IN THE AEDC PWT 4-FOOT TRANSONIC WIND TUNNEL
2335	151,783		IA140A/B	RESULTS OF EXPERIMENTAL INVESTIGATIONS IN THE MSFC TWT TO DETERMINE EFFECTS OF A MULTIPLE STING SUPPORT SYSTEM ON THE MATED VEHICLE AERODYNAMICS UTILIZING A 0.004 SCALE (74-OTS, 77-O) SHUTTLE VEHICLE 5 (IA140 A/B)
2336	167,375		LA145	INVESTIGATION OF THE HIGH ANGLE OF ATTACK AERODYNAMICS OF A SPACE SHUTTLE ORBITER(LARC .0098 SCALE MODEL) IN THE LARC UPWT AT MACH NUMBERS FROM 1.5 TO 4.5(LA145)
2337	151,786		OA236	A VERIFICATION STUDY OF THREE AMES RESEARCH CENTER PITOT-STATIC PROBES IN THE ROCKWELL INTERNATIONAL NAAL LOW SPEED WIND TUNNEL
2338	147,639		CS3	RESULTS OF THE LOW SPEED AEROELASTIC BUFFET TEST WITH A 0.046-SCALE MODEL (747-AX1322D-3/ORBITER 8-O) OF THE 747 CAM/ORBITER IN THE UNIVERSITY OF WASHINGTON WIND TUNNEL
2339			OS32	** DOCUMENTATION NOT COMPLETE **
2340	160,501	V-01	OH98	RESULTS OF TESTS ON A 0.0175-SCALE MODEL (60-O) OF THE SPACE SHUTTLE ORBITER TO DETERMINE RE-ENTRY MODE CONVECTIVE HEAT TRANSFER RATES ON THE UPPER WING SURFACE AND SSME NOZZLES IN THE AEDC VKF 'B' HYPERSONIC WIND TUNNEL (OH98)
2340	160,502	V-02	OH98	RESULTS OF TESTS ON A 0.0175-SCALE MODEL (60-O) OF THE SPACE SHUTTLE ORBITER TO DETERMINE RE-ENTRY MODE CONVECTIVE HEAT TRANSFER RATES ON THE UPPER WING SURFACE AND SSME NOZZLES IN THE AEDC VKF 'B' HYPERSONIC WIND TUNNEL (OH98)
2341	147,638		CS4/5	RESULTS OF TESTS CS4 AND CS5 TO INVESTIGATE DYNAMIC LOADS AND PRESSURES ON 0.03-SCALE MODELS (AX1319-3/4 AND 45-O) OF MATED 747 CAM AND SPACE SHUTTLE ORBITER IN THE BOEING TRANSONIC WIND TUNNEL

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2342	151,074		OH54B	RESULTS OF PHASE CHANGE PAINT HEAT TRANSFER TEST UTILIZING 0.040 SCALE 50 PERCENT FOREBODY MODELS (NO. 82-0) OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER IN THE AEDC VKF HYPERSONIC TUNNEL B
2343	160,849		LA85	PITOT PRESSURE SURVEYS ON THE LEEWARD SURFACE OF A 0.0045-SCALE MODEL ATP SHUTTLE ORBITER AT 30 DEGREES ANGLE OF ATTACK AND MACH 20 IN THE LARC 22 INCH HELIUM TUNNEL(LA85)
2344	151,788	V-01	LA77	TRANSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE (REMOTELY CONTROLLED ELEVON) MODEL 44-0 OF THE SPACE SHUTTLE ORBITER TESTED IN THE NASA/ARC 11-FOOT TRANSONIC WIND TUNNEL (LA77)
2344	151,789	V-02	LA77	TRANSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015-SCALE (REMOTELY CONTROLLED ELEVON) MODEL 44-0 OF THE SPACE SHUTTLE ORBITER TESTED IN THE NASA/ARC 11-FOOT TRANSONIC WIND TUNNEL (LA77)
2345	78195		SA21F	AERODYNAMIC ROLL CHARACTERISTICS OF A 0.00548 SCALE 146-INCH SOLID ROCKET BOOSTER REENTRY CONFIGURATION (MSFC MODEL NUMBER 486) OVER A PORTION OF THE REENTRY FLIGHT REGIME IN THE NASA/MSFC 14-INCH TRISONIC WIND TUNNEL
2346	151,385	V-01	IA142	RESULTS OF SRB SEPARATION TESTS USING THE 0.010-SCALE SSV MODEL 75-OTS IN THE AEDC VKF TUNNEL A
2346	151,386	V-02	IA142	RESULTS OF SRB SEPARATION TESTS USING THE 0.010-SCALE SSV MODEL 75-OTS IN THE AEDC VKF TUNNEL A
2346	151,387	V-03	IA142	RESULTS OF SRB SEPARATION TESTS USING THE 0.010-SCALE SSV MODEL 75-OTS IN THE AEDC VKF TUNNEL A
2347	160,482	V-01	CA15A	MATED AERODYNAMIC CHARACTERISTICS INVESTIGATION FOR 0.04-SCALE MODEL BOEING 747 CAM/ORBITER (MODEL AX1284 E-6) COMBINATION IN THE UNIVERSITY OF WASHINGTON AERONAUTICAL LABORATORY F. K. KIRSTEN WIND TUNNEL (CA15A)
2348	160,483	V-01	CA15B	MATED AERODYNAMIC CHARACTERISTICS INVESTIGATION FOR 0.04-SCALE MODELBOEING 747 CAM/ORBITER (MODEL AX1284 E-7) COMBINATION IN THE UNIVERSITY OF WASHINGTON AERONAUTICAL LABORATORY F.K. KIRSTEN WIND TUNNEL (CA15B)
2349	151,379		CA17	RESULTS OF TEST CA17 CONDUCTED IN THE UWAL LOW SPEED WIND TUNNEL USING THE MATED 0.04-SCALE 747 MODEL AX1284 AND 0.0405 SPACE SHUTTLEORBITER MODEL 43-0
2350	151,065		OH46	RESULTS OF PHASE CHANGE PAINT THERMAL MAPPING TEST OH46 USING THE 0.006-SCALE MODEL 90-0 IN THE NASA LARC VARIABLE DENSITY TUNNEL

CHRYSLER DATA MANAGEMENT SERVICES(DMS)
SPACE SHUTTLE WIND TUNNEL TEST PROGRAM
DATA REPORT DOCUMENTATION

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2351	160,853		OA238	RESULTS OF TEST OA238 USING THE SSV VEHICLE 102 0.10-SCALE FOREBODY MODEL NO. 99-0 IN THE NAAL LOW SPEED WIND TUNNEL TO INVESTIGATE AIR DATA SYSTEM CHARACTERISTICS
2352	151,383		LA91	A STUDY OF TRANSONIC BETA HYSTERESIS OF AN 0.015 SCALE MODEL 44-0 (SPACE SHUTTLE ORBITER TESTED IN THE NASA/LARC 8-FOOT TRANSONIC) PRESSURE TUNNEL (LA91)
2353	160,827		LA89	SUBSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.030-SCALE SPACE SHUTTLE ORBITER WITH TAILCONE (MODEL 201) TESTED IN THE NASA/ARC 11-FOOT TRANSONIC WIND TUNNEL (LA89)
2354	151,401	V-01	IA143	RESULTS OF SRB SEPARATION TESTS USING THE 0.010 SCALE SSV MODEL 75-OTS IN THE AEDC VKF TUNNEL A (IA143)
2354	151,402	V-02	IA143	RESULTS OF SRB SEPARATION TESTS USING THE 0.010 SCALE SSV MODEL 75-OTS IN THE AEDC VKF TUNNEL A (IA143)
2354	151,403	V-03	IA143	RESULTS OF SRB SEPARATION TESTS USING THE 0.010 SCALE SSV MODEL 75-OTS IN THE AEDC VKF TUNNEL A (IA143)
2354	151,404	V-04	IA143	RESULTS OF SRB SEPARATION TESTS USING THE 0.010 SCALE SSV MODEL 75-OTS IN THE AEDC VKF TUNNEL A (IA143)
2355	151,066		OH49A	RESULTS OF TEST OH49A OF THE .0175-SCALE SPACE SHUTTLE ORBITER MODEL22-O CONDUCTED IN THE AEDC VKF TUNNEL B TO DETERMINE AERO HEATING CHARACTERISTICS
2356	151,064		OH60	AERODYNAMIC HEATING RESULTS OBTAINED DURING TEST OH60 CONDUCTED IN THE AEDC VKF TUNNEL B USING THE 0.040-SCALE MODEL 83-0 OF THE SPACE SHUTTLE ORBITER FORWARD FIFTY PERCENT FUSELAGE
2357	167,655		IH68	RESULTS OF ASCENT AERODYNAMIC HEATING TESTS ON THE SPACE SHUTTLE ASCENT VEHICLE, AT MACH 5.3 AND 7.4 IN THE NASA/AMES 3.5-FOOT HWT, USING THE 0.0175-SCALE MODEL 60 OTS (IH68)
2358	151,067		OH50B	AERODYNAMIC HEATING RESULTS OBTAINED DURING TEST OH50B CONDUCTED IN THE AEDC VKF TUNNEL B USING THE 0.040-SCALE 83-0 OF THE SPACE SHUTTLE ORBITER FORWARD FIFTY PERCENT FUSELAGE
2359	151,405		OH66	RESULTS OF HEAT TRANSFER TESTING OF AN 0.025-SCALE MODEL (66-0) OF THE SPACE SHUTTLE ORBITER CONFIGURATION 140B IN THE CALSPAN HYPER- SONIC SHOCK TUNNEL (OH66)

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2360	160,521	V-01	OA221B/C	CALIBRATION TESTS OF THE SPACE SHUTTLE ORBITER PRIMARY AND ALTERNATE AIR DATA SYSTEMS USING A 0.10-SCALE ORBITER FOREBODY MODEL (99-0) IN THE NASA AMES RESEARCH CENTER 9 X 7 AND 8 X 7-FOOT LEGS OF THE UNITARY PLAN WIND TUNNEL (OA221B AND C)
2360	160,522	V-02	OA221B/C	CALIBRATION TESTS OF THE SPACE SHUTTLE ORBITER PRIMARY AND ALTERNATE AIR DATA SYSTEMS USING A 0.10-SCALE ORBITER FOREBODY MODEL (99-0) IN THE NASA AMES RESEARCH CENTER 9 X 7 AND 8 X 7-FOOT LEGS OF THE UNITARY PLAN WIND TUNNEL (OA221B AND C)
2361	151,370	V-01	OA163B	RESULTS OF A LANDING GEAR LOADS TEST USING A 0.0405-SCALE MODEL (16-0) OF THE SPACE SHUTTLE ORBITER IN THE ROCKWELL INTERNATIONAL NAAL WIND TUNNEL (OA163B)
2361	151,371	V-02	OA163B	RESULTS OF A LANDING GEAR LOADS TEST USING A 0.0405-SCALE MODEL (16-0) OF THE SPACE SHUTTLE ORBITER IN THE ROCKWELL INTERNATIONAL NAAL WIND TUNNEL (OA163B)
2362			LA92	** DOCUMENTATION NOT COMPLETE **
2363	151,057		OS7	RESULTS OF FLUTTER TEST OS7 OBTAINED USING THE 0.14-SCALE SPACE SHUTTLE ORBITER FIN/RUDDER MODEL NUMBER 55-0 IN THE NASA LARC 16-FOOT TRANSONIC DYNAMICS WIND TUNNEL
2364	160,527	V-01	OA145B	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-0) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145B)
2364	160,528	V-02	OA145B	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-0) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145B)
2364	160,529	V-03	OA145B	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-0) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145B)
2365	151,056		OS6	RESULTS OF FLUTTER TEST OS6 OBTAINED USING THE 0.14-SCALE WING/ELEVON MODEL (54-0) IN THE NASA LARC 16-FOOT TRANSONIC DYNAMICS WIND TUNNEL
2366	151,063		OH25B	HEAT TRANSFER PHASE CHANGE PAINT TESTS OF 0.0175-SCALE MODEL (NO. 56-0) OF THE ROCKWELL INTERNATIONAL SPACE SHUTTLE ORBITER IN THE AEDC TUNNEL B HYPERSONIC WIND TUNNEL

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2367	151,773		OH57A/B	RESULTS OF A HIGH ANGLE-OF-ATTACK AERO HEATING PRESSURE TEST ON A 0.0175-SCALE MODEL (92-0) OF THE OV-102 CONFIGURATION SPACE SHUTTLE ORBITER IN THE AEDC VKF TUNNEL B (OH57A/B)
2368	151,058		OH51	RESULTS OF PHASE CHANGE HEAT TRANSFER TEST OH51 USING 0.006-SCALE SPACE SHUTTLE ORBITER MODELS 46-0 AND 90-0 AND PARTIAL WING 0.0175-SCALE MODEL 64-0 IN THE LARC 31-INCH CFHT
2369	167,345		SA31F	AN AERODYNAMIC STATIC STABILITY WIND TUNNEL TEST OF A 0.00856 SCALE MODEL OF THE SPACE SHUTTLE 146 INCH DIAMETER SOLID ROCKET BOOSTER REENTRY CONFIGURATION (MSFC MODEL 487) IN THE NASA/MSFC HIGH REYNOLDS NUMBER WIND TUNNEL
2370	151,790	V-01	OA149B/C	RESULTS OF TEST USING A 0.030-SCALE PRESSURE LOADS SPACE SHUTTLE ORBITER MODEL (47-0) IN THE NASA/ARC UNITARY PLAN WIND TUNNEL
2370	151,791	V-02	OA149B/C	RESULTS OF TEST USING A 0.030-SCALE PRESSURE LOADS SPACE SHUTTLE ORBITER MODEL (47-0) IN THE NASA/ARC UNITARY PLAN WIND TUNNEL
2370	151,792	V-03	OA149B/C	RESULTS OF TEST USING A 0.030-SCALE PRESSURE LOADS SPACE SHUTTLE ORBITER MODEL (47-0) IN THE NASA/ARC UNITARY PLAN WIND TUNNEL
2371	151,408		OH78	RESULTS OF BASE HEATING TESTS ON A 0.04 SCALE SPACE SHUTTLE ORBITER BASE (MODEL 65-0) IN THE NASA/JSC THERMAL VACUUM CHAMBER A
2372	160,843		IH72	RESULTS OF HEAT TRANSFER TESTS OF A 0.0175-SCALE SPACE SHUTTLE INTEGRATED VEHICLE MODEL 60-OTS IN THE AEDC-VKF TUNNEL A (IH72)
2373	160,821		LA99	EFFECT OF TAILCONE CUT-OFF AND STING CONFIGURATION ON THE AERODYNAMIC CHARACTERISTICS OF A 0.030 SCALE(REMOTELY CONTROLLED ELEVON, BODYFLAP AND RUDDER) MODEL 201-0 ALT ORBITER TESTED IN THE NASA/LARC 8-FOOT TPT (LA99)
2374	167,372		LA82/LA103	INVESTIGATIONS IN THE CALSPAN 8-FOOT TRANSONIC WIND TUNNEL TO DETERMINE STING-TARE EFFECTS ON A MODIFIED 0.0165-SCALE SPACE SHUTTLE ORBITER MODEL WITH A TAILCONE (LA82/LA103)
2375	160,530		OA237	RESULTS OF AIR DATA SYSTEM CALIBRATION TEST USING THE 0.10-SCALE SPACE SHUTTLE ORBITER VEHICLE 102 FOREBODY MODEL 99-0 IN THE NASA 40 X 80-FOOT SUBSONIC WIND TUNNEL (OA237)
2376	151,779	V-01	OA149A	RESULTS OF TEST USING A 0.030-SCALE PRESSURE LOADS SPACE SHUTTLE ORBITER MODEL (47-0) IN THE NASA/ARC UNITARY PLAN WIND TUNNEL

CHRYSLER DATA MANAGEMENT SERVICES(DMS)
SPACE SHUTTLE WIND TUNNEL TEST PROGRAM
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DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2376	151,780	V-02	OA149A	RESULTS OF TEST USING A 0.030-SCALE PRESSURE LOADS SPACE SHUTTLE ORBITER MODEL (47-0) IN THE NASA/ARC UNITARY PLAN WIND TUNNEL
2376	151,781	V-03	OA149A	RESULTS OF TEST USING A 0.030-SCALE PRESSURE LOADS SPACE SHUTTLE ORBITER MODEL (47-0) IN THE NASA/ARC UNITARY PLAN WIND TUNNEL
2377	167,342	V-01	IA144	RESULTS OF TESTS OF THE 0.010 SCALE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/AMES RESEARCH CENTER 11X11 FOOT TRANSONIC WIND TUNNEL, MODEL 72-OTS TEST IA144
2377	167,343	V-02	IA144	RESULTS OF TESTS OF THE 0.010 SCALE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/AMES RESEARCH CENTER 11X11 FOOT TRANSONIC WIND TUNNEL, MODEL 72-OTS TEST IA144
2378	160,820		IA191	RESULTS OF AN INVESTIGATION OF STATIC AND DYNAMIC PRESSURE DISTRIBUTIONS ON EXTERNAL TANK PROTUBERANCES IN THE 11-FOOT LEG OF THE NASA/ARC UNITARY PLAN WIND TUNNEL (IA191)
2379			LA106	** DOCUMENTATION NOT COMPLETE **
2380	151,801	V-01	OA145A	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-0) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145A
2380	151,802	V-02	OA145A	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-0) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145A
2380	151,803	V-03	OA145A	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-0) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145A
2380	151,804	V-04	OA145A	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-0) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145A
2380	151,805	V-05	OA145A	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-0) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145A

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2380	151,806	V-06	OA145A	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-O) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145A
2381			LA107	
2382	151,382		OH8/IA109	RESULTS OF EXPERIMENTAL TESTS IN THE NASA/MSFC IMPULSE BASE FLOW FACILITY ON A SPACE SHUTTLE .04 SCALE ORBITER (MODEL 25-O) TO DETERMINE SECOND STAGE ASCENT BASE HEATING RATES AND PRESSURE DISTRIBUTION
2383			LA93	** DOCUMENTATION NOT COMPLETE **
2384	151,412	V-01	IA148	RESULTS OF RCS JET PLUME INTERACTION TESTS USING A 0.0125-SCALE MODEL (70-OT) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE AEDC VKF TUNNEL %B% (IA148)
2384	151,413	V-02	IA148	RESULTS OF RCS JET PLUME INTERACTION TESTS USING A 0.0125-SCALE MODEL (70-OT) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE AEDC VKF TUNNEL %B% (IA148)
2385	151,366		OH15	RESULTS OF TESTS ON A 0.111-SCALE SPACE SHUTTLE VEHICLE SIMULATED ELEVON/WING GAP HEAT TRANSFER MODEL (53-O) IN THE AMES RESEARCH CENTER 3.5-FOOT HWT
2386	151,368		OH44	RESULTS OF TESTS ON A 0.111-SCALE SPACE SHUTTLE VEHICLE SIMULATED ELEVON/ELEVON GAP HEAT TRANSFER MODEL (53-O) IN THE AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL
2387	TASK CANCELLED		LA104	
2388	167,676		OH84A	RESULTS OF WIND TUNNEL TESTS OF THIN-SKIN THERMOCOUPLE MODELS 83-O (0.04-SCALE) AND 60-O (0.0175-SCALE) OF THE SPACE SHUTTLE ORBITER IN THE AEDC VKF HYPERSONIC WIND TUNNEL B (OH84A)
2389	160,810	V-01	OA145C	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-O) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145C)
2389	160,811	V-02	OA145C	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-O) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145C)
2389	160,812	V-03	OA145C	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER VEHICLE 102 AERO CHARACTERISTICS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL (39-O) IN THE AMES RESEARCH CENTER UNITARY WIND TUNNEL(OA145C)

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2390	160,481		LA101	LOW SUPERSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.0015-SCALE (REMOTELY CONTROLLED ELEVON) MODEL 44-0 SPACE SHUTTLE ORRITER TESTED IN THE NASA/LARC 4 FOOT UPWT (LEG 1) (LA101)
2391	167,346		IA244	RESULTS OF TESTS OF THE 0.10 SCALE SPACE SHUTTLE INTEGRATED VEHICLE IN THE LANGLEY RESEARCH CENTER 8-FOOT TRANSONIC PRESSURE TUNNEL, MODEL 72-OTS TEST IA244
2392	151,389		OA250	GROUND PROXIMITY TESTS OF THE 0.03-SCALE MODEL (45-0) SPACE SHUTTLE ORBITER IN THE ROCKWELL INTERNATIONAL NAAL LOW SPEED WIND TUNNEL
2393	167,679	V-01	IH51A	RESULTS OF SSV INTERFERENCE HEATING TESTS ON A 0.04-SCALE THIN-SKIN THERMOCOUPLE MODEL (58-OT) UTILIZING A SIMULATED EXTERNAL TANK & ORBITER FOREBODY IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (IH51A)
2393	167,680	V-02	IH51A	RESULTS OF SSV INTERFERENCE HEATING TESTS ON A 0.04-SCALE THIN-SKIN THERMOCOUPLE MODEL (58-OT) UTILIZING A SIMULATED EXTERNAL TANK & ORBITER FOREBODY IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (IH51A)
2393	167,681	V-03	IH51A	RESULTS OF SSV INTERFERENCE HEATING TESTS ON A 0.04-SCALE THIN-SKIN THERMOCOUPLE MODEL (58-OT) UTILIZING A SIMULATED EXTERNAL TANK & ORBITER FOREBODY IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (IH51A)
2393	167,682	V-04	IH51A	RESULTS OF SSV INTERFERENCE HEATING TESTS ON A 0.04-SCALE THIN-SKIN THERMOCOUPLE MODEL (58-OT) UTILIZING A SIMULATED EXTERNAL TANK & ORBITER FOREBODY IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (IH51A)
2394			LA109	** DOCUMENTATION NOT COMPLETE **
2395	151,394		LA111	EFFECT OF SILTS POD ON THE TRANSONIC AERODYNAMIC CHARACTERISTICS OF A 0.015-SCALE SHUTTLE ORBITER MODEL (44-0) TESTED IN THE NASA/LARC 8-FOOT TPT
2396	151,393		LA110	EFFECT OF SILTS POD ON THE LOW SUPERSONIC AERODYNAMIC CHARACTERISTICS OF A 0.015-SCALE SHUTTLE ORBITER MODEL (44-0) TESTED IN THE NASA/LARC 4-FOOT UPWT (LEG 1)
2397	167,347		LA113	RESULTS OF WIND TUNNEL TESTS ON A 0.010 SCALE MODEL (72-OTS) ROCKWELL SPACE SHUTTLE VEHICLE IN THE LARC 8-FOOT TRANSONIC PRESSURE TUNNEL (LA113)
2398	160,850	V-01	IA105A	RESULTS OF TESTS USING A 0.03 SCALE MODEL (47-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC 16 FOOT TRANSONIC PROPULSION WIND TUNNEL (IA105A)

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2398	160,851	V-02	IA105A	RESULTS OF TESTS USING A 0.03 SCALE MODEL (47-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC 16 FOOT TRANSONIC PROPULSION WIND TUNNEL (IA105A)
2398	160,852	V-03	IA105A	RESULTS OF TESTS USING A 0.03 SCALE MODEL (47-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC 16 FOOT TRANSONIC PROPULSION WIND TUNNEL (IA105A)
2399	151,388		LA114	EFFECT OF SILTS POD ON THE HIGH SUPERSONIC AERODYNAMIC CHARACTERISTICS OF A 0.015-SCALE SHUTTLE ORBITER MODEL (44-0) TESTED IN THE NASA/LARC 4-FOOT UPWT (LEG 2)
2400	160,518		OA234	RESULTS OF SSV ORBITER AIR DATA SYSTEM CALIBRATION TEST USING THE 0.10-SCALE ORBITER FOREBODY MODEL 99-0 IN THE NASA/LEWIS 10 X 10-FOOT SUPERSONIC WIND TUNNEL (OA234)
2401	151,395		IS1A/B/C/OS3	AERONOISE TEST RESULTS USING A 0.040-SCALE SPACE SHUTTLE VEHICLE CONFIGURATION 2A MODEL (11-OTS) IN THE AMES RESEARCH CENTER UNITARY PLAN WIND TUNNELS
2402	151,763		OA223	SYSTEM CHECKOUT OF THE 0.05-SCALE SPACE SHUTTLE VEHICLE ORBITER 102 MODEL (39-0) IN THE NAAL LOW SPEED WIND TUNNEL(OA223)
2403	160,515	V-01	IA156A	RESULTS OF TESTS USING A 0.02-SCALE MODEL (89-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC 16-FOOT TRANSONIC PROPULSION WIND TUNNEL (IA156A)
2403	160,516	V-02	IA156A	RESULTS OF TESTS USING A 0.02-SCALE MODEL (89-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC 16-FOOT TRANSONIC PROPULSION WIND TUNNEL (IA156A)
2403	160,517	V-03	IA156A	RESULTS OF TESTS USING A 0.02-SCALE MODEL (89-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC 16-FOOT TRANSONIC PROPULSION WIND TUNNEL (IA156A)
2404	160,510	V-01	IA119	RESULTS OF TESTS USING A 0.020-SCALE MODEL (88-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE JET PLUME IN THE NASA/ARC UPWT 11 X 11-FOOT LEC (TEST IA119)
2404	160,511	V-02	IA119	RESULTS OF TESTS USING A 0.020-SCALE MODEL (88-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE JET PLUME IN THE NASA/ARC UPWT 11 X 11-FOOT LEC (TEST IA119)

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2404	160,512	V-03	IA119	RESULTS OF TESTS USING A 0.020-SCALE MODEL (88-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE JET PLUME IN THE NASA/ARC UPWT 11 X 11-FOOT LEC (TEST IA119)
2404	160,513	V-04	IA119	RESULTS OF TESTS USING A 0.020-SCALE MODEL (88-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE JET PLUME IN THE NASA/ARC UPWT 11 X 11-FOOT LEC (TEST IA119)
2405	151,756	V-01	OA101	RESULTS OF A LOW SPEED APPROACH AND LANDING EXPERIMENTAL INVESTIGA- TION OF A 0.050-SCALE SPACE SHUTTLE ORBITER MODEL (39-O) IN THE NASA/AMES RESEARCH CENTER'S 12-FOOT PRESSURE WIND TUNNEL (OA101)
2405	151,757	V-02	OA101	RESULTS OF A LOW SPEED APPROACH AND LANDING EXPERIMENTAL INVESTIGA- TION OF A 0.050-SCALE SPACE SHUTTLE ORBITER MODEL (39-O) IN THE NASA/AMES RESEARCH CENTER'S 12-FOOT PRESSURE WIND TUNNEL (OA101)
2405	151,758	V-03	OA101	RESULTS OF A LOW SPEED APPROACH AND LANDING EXPERIMENTAL INVESTIGA- TION OF A 0.050-SCALE SPACE SHUTTLE ORBITER MODEL (39-O) IN THE NASA/AMES RESEARCH CENTER'S 12-FOOT PRESSURE WIND TUNNEL (OA101)
2405	151,759	V-04	OA101	RESULTS OF A LOW SPEED APPROACH AND LANDING EXPERIMENTAL INVESTIGA- TION OF A 0.050-SCALE SPACE SHUTTLE ORBITER MODEL (39-O) IN THE NASA/AMES RESEARCH CENTER'S 12-FOOT PRESSURE WIND TUNNEL (OA101)
2405	151,760	V-05	OA101	RESULTS OF A LOW SPEED APPROACH AND LANDING EXPERIMENTAL INVESTIGA- TION OF A 0.050-SCALE SPACE SHUTTLE ORBITER MODEL (39-O) IN THE NASA/AMES RESEARCH CENTER'S 12-FOOT PRESSURE WIND TUNNEL (OA101)
2405	151,761	V-06	OA101	RESULTS OF A LOW SPEED APPROACH AND LANDING EXPERIMENTAL INVESTIGA- TION OF A 0.050-SCALE SPACE SHUTTLE ORBITER MODEL (39-O) IN THE NASA/AMES RESEARCH CENTER'S 12-FOOT PRESSURE WIND TUNNEL (OA101)
2406	167,348		IA181	RESULTS OF AN EXPERIMENTAL INVESTIGATION IN THE NASA/MSFC 14-INCH TRISONIC WIND TUNNEL ON A .004-SCALE MODEL (74-OTS) SSLV TO DETERMINE INFLUENCE OF ORBITER AND SRB'S ON TEH EXTERNAL TANK NOSE PRESSURE DISTRIBUTION (IA181)
2407	167,374		IH73	RESULTS OF M=5.3 HEAT TRANSFER TESTS ON THE SECOND STAGE SPACE SHUTTLE CONFIGURATION AT RTLS ABORT MISSION PROFILE CONDITIONS USING THE 0.006 SCALE MODEL 50-O & 41-T IN THE NASA/ARC 3.5-FOOT HWT (IH73)

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2408	160,498	V-01	IA156B	RESULTS OF TESTS USING A 0.02-SCALE MODEL (89-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/AMES RESEARCH CENTER 9X7 FOOT SUPERSONIC WIND TUNNEL (IA156B)
2408	160,499	V-02	IA156B	RESULTS OF TESTS USING A 0.02-SCALE MODEL (89-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/AMES RESEARCH CENTER 9X7 FOOT SUPERSONIC WIND TUNNEL (IA156B)
2408	160,500	V-03	IA156B	RESULTS OF TESTS USING A 0.02-SCALE MODEL (89-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/AMES RESEARCH CENTER 9X7 FOOT SUPERSONIC WIND TUNNEL (IA156B)
2409	160,842		LA115	ADDITIONAL TRANSONIC STABILITY AND CONTROL CHARACTERISTICS OF A 0.015 SCALE(REMOTELY CONTROLLED ELEVON) MODEL 44-0 SPACE SHUTTLE ORBITER TESTED IN THE NASA/LARC 8-FOOT TPT (LA115)
2410	151,777		OH56	RESULTS OF THE NASA/RI ORBITER WING TIP HEATING TEST WITH THE 0.08-SCALE ORBITER WING MODEL (91-0) IN THE AEDC VKF B HYPERSONIC WIND T UNNEL (OH56)
2411			LA116	** DOCUMENTATION NOT COMPLETE **
2412	167,386	V-01	IH90	RESULTS OF HEAT TRANSFER TESTS ON THE SPACE SHUTTLE INTEGRATED VEHICLE, UNDER ASCENT CONDITIONS, USING THE 0.0175-SCALE 60-OTS MODEL IN THE NASA/ARC 3.5-FOOT HWT (IH-90)
2412	167,387	V-02	IH90	RESULTS OF HEAT TRANSFER TESTS ON THE SPACE SHUTTLE INTEGRATED VEHICLE, UNDER ASCENT CONDITIONS, USING THE 0.0175-SCALE 60-OTS MODEL IN THE NASA/ARC 3.5-FOOT HWT (IH-90)
2413	160,858	V-01	IA105B	RESULTS OF TESTS USING A 0.03 SCALE MODEL (47-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/ARC 9X7 FOOT SUPERSONIC WIND TUNNEL (IA105B)
2413	160,859	V-02	IA105B	RESULTS OF TESTS USING A 0.03 SCALE MODEL (47-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/ARC 9X7 FOOT SUPERSONIC WIND TUNNEL (IA105B)
2414	160,484	V-01	OA232	CALIBRATION TESTS OF THE SPACE SHUTTLE AIR DATA SYSTEM USING A 0.10-SCALE ORBITER FOREBODY MODEL (99-0) IN THE AEDC 16T PROPULSION WIND TUNNEL (OA232)
2414	160,485	V-02	OA232	CALIBRATION TESTS OF THE SPACE SHUTTLE AIR DATA SYSTEM USING A 0.10-SCALE ORBITER FOREBODY MODEL (99-0) IN THE AEDC 16T PROPULSION WIND TUNNEL (OA232)

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2415	151,784	V-01	OA208/209	RESULTS OF TESTS USING A 0.02-SCALE MODEL (105-0) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER VON KARMAN FACILITY SUPERSONIC TUNNEL A (OA209) AND HYPERSONIC TUNNEL B (OA208/209)
2415	151,785	V-02	OA208/209	RESULTS OF TESTS USING A 0.02-SCALE MODEL (105-0) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER VON KARMAN FACILITY SUPERSONIC TUNNEL A (OA209) AND HYPERSONIC TUNNEL B (OA208/209)
2416	160,824		IA603	RESULTS OF TESTS IN THE NASA/MSFC 14-INCH TRISONIC WIND TUNNEL ON A .004 SCALE MODEL (74-OTS) THRUST AUGMENTED SPACE SHUTTLE INTEGRATED VEHICLE (IA603)
2417	151,770		OH58	RESULTS OF AEROTHERMODYNAMIC HEAT TRANSFER TESTS ON A 0.03-SCALE MODEL (93-0) SIMULATING THE ELEVON/ELEVON GAP AND ELEVON/FUSELAGE INTERFACE REGIONS OF THE SS ORBITER IN THE ARC 3.5HWT.
2418	151,414		IH100	RESULTS OF TESTS OF A DEVELOPMENT FLIGHT INSTRUMENTATION GAS TEMPERATURE PROBE IN THE AMES RESEARCH CENTER 3.5 FT. HYPERSONIC WIND TUNNEL (IH100)
2419	151,762		OA270B/C	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER AERO-CHARACTERISTICS AND EXAMINE TRANSONIC BLOCKAGE AND SHOCK REFLECTION EFFECTS UTILIZING .02-SCALE HI-FIDELITY MODELS 104-0 AND 105-0 IN THE LANGLEYRESEARCH CENTER 16-FT. TRANSONIC WIND TUNNEL OA270B/C
2420	167,385		OH103A	RESULTS OF TESTS ON A 0.04-SCALE SPACE SHUTTLE ORBITER FOREBODY MODEL.(83-0) IN THE AEDC VKF HYPERSONIC WIND TUNNEL 'B' TO OBTAIN AERODYNAMIC HEATING DISTRIBUTION ON LOWER FUSELAGE AND RCS NOZZLE AREAS (OH103A)
2421	160,495	V-01	OA251B/C	CALIBRATION TESTS OF THE SPACE SHUTTLE ORBITER AIR DATA SYSTEM USING A 0.10-SCALE ORBITER FOREBODY MODEL (99.0) IN THE NASA AMES RESEARCH CENTER 9 X 7 AND 8 X 7-FOOT LEGS OF THE UNITARY PLAN WIND TUNNEL (OA251B AND C)
2421	160,496	V-02	OA251B/C	CALIBRATION TESTS OF THE SPACE SHUTTLE ORBITER AIR DATA SYSTEM USING A 0.10-SCALE ORBITER FOREBODY MODEL (99.0) IN THE NASA AMES RESEARCH CENTER 9 X 7 AND 8 X 7-FOOT LEGS OF THE UNITARY PLAN WIND TUNNEL (OA251B AND C)
2422	151,767		FH15	RESULTS OF THIN SKIN THERMOCOUPLE TESTS CONDUCTED IN THE AEDC VKF TUNNEL A TO DETERMINE HEAT TRANSFER RATES ON A .0275 SCALE SSV ET FOREBODY (FH15)
2423	151,768		FH16	RESULTS OF THIN SKIN THERMOCOUPLE TESTS CONDUCTED IN THE NASA/ARC 3.5 FT. HYPERSONIC WIND TUNNEL TO DETERMINE HEAT TRANSFER RATES ON A .0275 SCALE SSV ET FOREBODY(FH16)

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2424	160,506	V-01	OA126A,B,C	RESULTS OF TESTS ON THE EFFECTS OF AEROELASTICITY OF THE SPACE SHUTTLE ORBITER VERTICAL TAIL USING A 0.03-SCALE MODEL (47-0) IN THE NASA AMES UNITARY WIND TUNNELS (OA126A/B)
2424	160,507	V-02	OA126A,B,C	RESULTS OF TESTS ON THE EFFECTS OF AEROELASTICITY OF THE SPACE SHUTTLE ORBITER VERTICAL TAIL USING A 0.03-SCALE MODEL (47-0) IN THE NASA AMES UNITARY WIND TUNNELS (OA126A/B)
2424	160,508	V-03	OA126A,B,C	RESULTS OF TESTS ON THE EFFECTS OF AEROELASTICITY OF THE SPACE SHUTTLE ORBITER VERTICAL TAIL USING A 0.03-SCALE MODEL (47-0) IN THE NASA AMES UNITARY WIND TUNNELS (OA126ABC)
2425			LA117	** DOCUMENTATION NOT COMPLETE **
2426	TP1186		LA124	A WIND TUNNEL STUDY OF THE APPLICABILITY OF FAR-FIELD SONIC-BOOM THEORY TO THE SPACE SHUTTLE ORBITER
2427	167,675		OH103B	RESULTS OF TESTS OF A 0.0175-SCALE THIN-SKIN THERMOCOUPLE WIND TUNNEL MODEL (60-0) OF THE SPACE SHUTTLE ORBITER TO DETERMINE EFFECTS OF SURFACE ROUGHNESS IN THE AEDC VKF HYPERSONIC WIND TUNNEL B (OH103B)
2428	160,523	V-01	IH11	WIND TUNNEL TESTS OF THE 0.035-SCALE INTEGRATED SPACE SHUTTLE VEHICLE 84-OTS IN THE NASA/LEWIS 10 X 10-FOOT SUPERSONIC WIND TUNNEL (IH11)
2428	160,524	V-02	IH11	WIND TUNNEL TESTS OF THE 0.035-SCALE INTEGRATED SPACE SHUTTLE VEHICLE 84-OTS IN THE NASA/LEWIS 10 X 10-FOOT SUPERSONIC WIND TUNNEL (IH11)
2428	160,525	V-03	IH11	WIND TUNNEL TESTS OF THE 0.035-SCALE INTEGRATED SPACE SHUTTLE VEHICLE 84-OTS IN THE NASA/LEWIS 10 X 10-FOOT SUPERSONIC WIND TUNNEL (IH11)
2428	160,526	V-04	IH11	WIND TUNNEL TESTS OF THE 0.035-SCALE INTEGRATED SPACE SHUTTLE VEHICLE 84-OTS IN THE NASA/LEWIS 10 X 10-FOOT SUPERSONIC WIND TUNNEL (IH11)
2429	167,353		IH51B	THIN SKIN HEAT TRANSFER TESTS OF A SIMULATED SPACE SHUTTLE 0.04 SCALE SOLID ROCKET BOOSTER/ET MODEL (58-TS) IN THE NASA/ARC 3.5 FOOT HYPERSONIC WIND TUNNEL (IH51B)
2430	160,817	V-01	OA270A	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER AERO-CHARACTERISTICS AND EXAMINE TRANSONIC BLOCKAGE AND SHOCK REFLECTION EFFECTS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL(39-0) IN THELANGLEY RESEARCH CENTER 16-FT. TRANSONIC WIND TUNNEL OA270A

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2430	160,818	V-02	OA270A	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER AERO-CHARACTERISTICS AND EXAMINE TRANSONIC BLOCKAGE AND SHOCK REFLECTION EFFECTS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL(39-O) IN THELANGLEY RESEARCH CENTER 16-FT. TRANSONIC WIND TUNNEL OA270A
2430	160,819	V-03	OA270A	RESULTS OF AN INVESTIGATION TO VERIFY SHUTTLE ORBITER AERO-CHARACTERISTICS AND EXAMINE TRANSONIC BLOCKAGE AND SHOCK REFLECTION EFFECTS UTILIZING AN .05-SCALE HI-FIDELITY REMOTE CONTROL MODEL(39-O) IN THELANGLEY RESEARCH CENTER 16-FT. TRANSONIC WIND TUNNEL OA270A
2431	151,793	V-01	IH85	TEST RESULTS FROM THE NASA/ROCKELL INTERNATIONAL SPACE SHUTTLE INTEGRATED VEHICLE TEST USING A 0.0175-SCALE MODEL (60-OTS) CONDUCTED IN THE AEDC-VKF TUNNEL A (IH85)
2431	151,794	V-02	IH85	TEST RESULTS FROM THE NASA/ROCKELL INTERNATIONAL SPACE SHUTTLE INTEGRATED VEHICLE TEST USING A 0.0175-SCALE MODEL (60-OTS) CONDUCTED IN THE AEDC-VKF TUNNEL A (IH85)
2431	151,795	V-03	IH85	TEST RESULTS FROM THE NASA/ROCKELL INTERNATIONAL SPACE SHUTTLE INTEGRATED VEHICLE TEST USING A 0.0175-SCALE MODEL (60-OTS) CONDUCTED IN THE AEDC-VKF TUNNEL A (IH85)
2431	151,796	V-04	IH85	TEST RESULTS FROM THE NASA/ROCKELL INTERNATIONAL SPACE SHUTTLE INTEGRATED VEHICLE TEST USING A 0.0175-SCALE MODEL (60-OTS) CONDUCTED IN THE AEDC-VKF TUNNEL A (IH85)
2431	151,797	V-05	IH85	TEST RESULTS FROM THE NASA/ROCKELL INTERNATIONAL SPACE SHUTTLE INTEGRATED VEHICLE TEST USING A 0.0175-SCALE MODEL (60-OTS) CONDUCTED IN THE AEDC-VKF TUNNEL A (IH85)
2431	151,798	V-06	IH85	TEST RESULTS FROM THE NASA/ROCKELL INTERNATIONAL SPACE SHUTTLE INTEGRATED VEHICLE TEST USING A 0.0175-SCALE MODEL (60-OTS) CONDUCTED IN THE AEDC-VKF TUNNEL A (IH85)
2431	151,799	V-07	IH85	TEST RESULTS FROM THE NASA/ROCKELL INTERNATIONAL SPACE SHUTTLE INTEGRATED VEHICLE TEST USING A 0.0175-SCALE MODEL (60-OTS) CONDUCTED IN THE AEDC-VKF TUNNEL A (IH85)
2431	151,800	V-08	IH85	TEST RESULTS FROM THE NASA/ROCKELL INTERNATIONAL SPACE SHUTTLE INTEGRATED VEHICLE TEST USING A 0.0175-SCALE MODEL (60-OTS) CONDUCTED IN THE AEDC-VKF TUNNEL A (IH85)

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DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2432	160,845		LA125	INVESTIGATION OF LONGITUDINAL AND LATERAL-DIRECTIONAL AERODYNAMIC CHARACTERISTICS FOR A 2 PERCENT (MODEL 105-0) SPACE SHUTTLE ORBITER (VEHICLE 102) IN THE LARC UPWT AT MACH NUMBERS FROM 2.5 TO 4.5 (LA125)
2433	151,764		OA171	RESULTS OF TESTS USING A 0.020-SCALE MODEL (105-0) OF THE SPACE SHUTTLE VEHICLE ORBITER IN THE NAVAL SURFACE WEAPONS CENTER HYPERVELOCITY TUNNEL 9 (OA171)
2434	151,782		OA129	RESULTS OF TESTS ON THE EFFECTS OF AEROELASTICITY OF THE SPACE SHUTTLE ORBITER VERTICAL TAIL USING A 0.03-SCALE MODEL (47-0) IN THE AEDC-16T PROPULSION WIND TUNNEL (OA129)
2435	151,415		IH39	BASE PRESSURE AND HEAT TRANSFER TESTS OF THE 0.0225-SCALE SPACE SHUTTLE PLUME SIMULATION MODEL (19-OTS) IN THE NASA-LEWIS RESEARCH CENTER 10X10-FOOT SUPERSONIC WIND TUNNEL (TEST IH39)
2436	TM-X72661	V-06	LA126	SPACE SHUTTLE ORBITER TRIMMED CENTER OF GRAVITY EXTENSION STUDY VOL UME VI--SYSTEM DESIGN STUDIES
2437	151,766		FA25	RESULTS OF TRANSONIC TESTS IN THE NASA/MSFC 14-INCH TRISONIC WIND TUNNEL ON A 0.004 SCALE MODEL (74-OTS) SPACE SHUTTLE LAUNCH VEHICLE (FA25)
2438	160,855	V-01	IA138	RESULTS OF AN EXPERIMENTAL INVESTIGATION TO DETERMINE ORBITER AND SOLID ROCKET BOOSTER JET PLUME INDUCED EFFECTS UTILIZING A .01-SCALE INTEGRATED VEHICLE SPACE SHUTTLE MODEL (75-OTS) IN THE NASA/ARC 9X7 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL
2438	160,856	V-02	IA138	RESULTS OF AN EXPERIMENTAL INVESTIGATION TO DETERMINE ORBITER AND SOLID ROCKET BOOSTER JET PLUME INDUCED EFFECTS UTILIZING A .01-SCALE INTEGRATED VEHICLE SPACE SHUTTLE MODEL (75-OTS) IN THE NASA/ARC 9X7 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL
2438	160,857	V-03	IA138	RESULTS OF AN EXPERIMENTAL INVESTIGATION TO DETERMINE ORBITER AND SOLID ROCKET BOOSTER JET PLUME INDUCED EFFECTS UTILIZING A .01-SCALE INTEGRATED VEHICLE SPACE SHUTTLE MODEL (75-OTS) IN THE NASA/ARC 9X7 FOOT LEG OF THE UNITARY PLAN WIND TUNNEL
2439	167,673		IA182	RESULTS OF TESTS USING A 0.03-SCALE MODEL (47-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC 16-FOOT TRANSONIC PROPULSION WIND TUNNEL (IA182)

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2440	151,765		IH83	BASE PRESSURE AND HEAT TRANSFER TESTS OF THE 0.0225-SCALE SPACE SHUTTLE PLUME SIMULATION MODEL (19-OTS) IN YAWED FLIGHT CONDITIONS IN THE NASA-LEWIS 10X10-FOOT SUPERSONIC WIND TUNNEL
2441			LA127	** DOCUMENTATION NOT COMPLETE **
2442			LA128	** DOCUMENTATION NOT COMPLETE **
2443	151,769		OH79	PRESSURE AND HEAT TRANSFER TESTS OF THE 0.040-SCALE SPACE SHUTTLE ORBITER BASE HEATING MODEL (65-O) IN THE JSC THERMAL VACUUM CHAMBER A.
2444	160,488	V-01	IA183	RESULTS OF TESTS USING A 0.02-SCALE MODEL (89-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC 16-FOOT TRANSONIC PROPULSION WINDTUNNEL (IA183)
2444	160,489	V-02	IA183	RESULTS OF TESTS USING A 0.02-SCALE MODEL (89-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE AEDC 16-FOOT TRANSONIC PROPULSION WINDTUNNEL (IA183)
2445	167,652	V-01	OA146	RESULTS OF A WIND TUNNEL PRESSURE LOADS TEST OF THE 0.03-SCALE SPACE SHUTTLE ORBITER (MODEL 47-O) IN THE 8X7-FOOT LEG OF THE NASA/ARC UNITARY PLAN WIND TUNNEL (OA146)
2[illegible]	167,653	V-02	OA146	RESULTS OF A WIND TUNNEL PRESSURE LOADS TEST OF THE 0.03-SCALE SPACE SHUTTLE ORBITER (MODEL 47-O) IN THE 8X7-FOOT LEG OF THE NASA/ARC UNITARY PLAN WIND TUNNEL (OA146)
2446			LA122	** DOCUMENTATION NOT COMPLETE **
2447			OS52	** DOCUMENTATION NOT COMPLETE **
2448	160,519	V-01	IH51C	SPACE SHUTTLE THIN SKIN HEAT TRANSFER TESTS OF SIMULATED LARGE SCALEPROTUBERANCES AND HALF SCALE TILE ON FLAT PLATE MODEL 58-OTS IN THE NASA AMES RESEARCH CENTER 3.5-FT HYPERSONIC WIND TUNNEL (IH51C)
2448	160,520	V-02	IH51C	SPACE SHUTTLE THIN SKIN HEAT TRANSFER TESTS OF SIMULATED LARGE SCALEPROTUBERANCES AND HALF SCALE TILE ON FLAT PLATE MODEL 58-OTS IN THE NASA AMES RESEARCH CENTER 3.5-FT HYPERSONIC WIND TUNNEL (IH51C)
2449	160,497		IA132	RESULTS OF SHUTTLE TRANSPORTATION SYSTEM ASCENT AIR DATA SYSTEM CALIBRATION TEST USING THE 0.07-SCALE EXTERNAL TANK FOREBODY MODEL (68-T) IN THE AEDC PWT 16-FOOT TRANSONIC WIND TUNNEL (IA132)

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2450	151,774		OS4A/OS4B/OS12	EXPERIMENTAL RESULTS OF TESTS TO DETERMINE THE EFFECTS OF ORBITER THERMAL PROTECTION SUBSYSTEM (TPS) TILES ON PANEL FLUTTER CONDUCTED INTHE ARC 2X2 TWT.
2451	151,772		OH90A/MA29	RESULTS OF BOUNDARY LAYER TRANSITION TESTS OF THE 0.025-SCALE RIGHT-HAND WING AND TRUNCATED AFT FUSELAGE MODEL (94-0) IN THE AEDC HWTB.
2452	167,383		IH99	RESULTS OF HEAT TRANSFER TESTS ON THE SPACE SHUTTLE FORWARD SRB SECTION AT ASCENT CONDITIONS USING THE 0.10-SCALE MODEL 98-S IN THE NASA/AMES 3.5-FOOT HWT (IH99)
2453	151,776		IH75	BASE PRESSURE AND HEAT TRANSFER TESTS OF THE 0.0225-SCALE SPACE SHUTTLE PLUME SIMULATION MODEL (19-OTS) IN THE NASA/CALSPAN LUDWIEG TUBEWIND TUNNEL
2454	TM-X72661	V-03	LA57	IMPACT OF RETROFITS FOR CENTER-OF-GRAVITY EXTENSION ON ORBITER THERMAL PROTECTION SYSTEM
2455	151,778		OH102A	RESULTS OF FLOW ANGULARITY TESTS ON A 0.0175-SCALE SPACE SHUTTLE ORBITER MODEL (56-0) ON THE AEDC VKF B HYPERSONIC WIND TUNNEL (OH102A)
2456	160,486	V-01	IA184	RESULTS OF TESTS USING A 0.03-SCALE MODEL (47-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/AMES RESEARCH CENTER 9X7 FOOT SUPERSONIC WIND TUNNEL (IA184)
2456	160,487	V-02	IA184	RESULTS OF TESTS USING A 0.03-SCALE MODEL (47-OTS) OF THE SPACE SHUTTLE INTEGRATED VEHICLE IN THE NASA/AMES RESEARCH CENTER 9X7 FOOT SUPERSONIC WIND TUNNEL (IA184)
2457	160,813		IA180	RESULTS OF SHUTTLE TRANSPORTATION SYSTEM ASCENT AIR DATA SYSTEM HIGH SUPERSONIC CALIBRATION TEST USING THE 0.07-SCALE EXTERNAL OXYGEN HYDROGEN TANK FOREBODY MODEL (68-T) IN THE UNITARY PLAN HIGH SPEED LEG OF THE LARC 4X4 WIND TUNNEL (IA180)
2458	167,668		OS36/37	SPACE SHUTTLE HRSI TILE TESTS OS36 AND OS37 IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT AND 9X7-FOOT WIND TUNNELS USING TEST FIXTURES 96-O AND 81-O (OS36/37)
2459	167,685	V-01	OA310A/OA310B/OA31	** DOCUMENTATION NOT COMPLETE **
2459	167,686	V-02	OA310A/OA310B/OA31	** DOCUMENTATION NOT COMPLETE **
2460			FA27	** DOCUMENTATION NOT COMPLETE **

DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2461	167,677		IH51D	SPACE SHUTTLE TESTS OF TURBULENT BOUNDARY LAYER HEATING EFFECTS ON HALF-SCALE TILE SIMULATION USING MODEL 58-O IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (IH51D)
2462	167,370	V-01	IA131B/C	RESULTS OF SUPERSONIC ASCENT AIR DATA SYSTEM CALIBRATION TESTS IA131B/C USING THE 0.07-SCALE EXTERNAL TANK FOREBODY MODEL 68-T IN THE ARC 9X7 AND 8X7 LEGS OF THE AMES UNITARY PLAN WIND TUNNEL
2462	167,371	V-02	IA131B/C	RESULTS OF SUPERSONIC ASCENT AIR DATA SYSTEM CALIBRATION TESTS IA131B/C USING THE 0.07-SCALE EXTERNAL TANK FOREBODY MODEL 68-T IN THE ARC 9X7 AND 8X7 LEGS OF THE AMES UNITARY PLAN WIND TUNNEL
2463	167,672		OS41/OS42/OS45	SPACE SHUTTLE LRSI TPS TILE TESTS OS41.OS42 AND OS45 IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT WIND TUNNEL USING MODEL 107-O (OS41.OS42 AND OS45)
2464	160,828	V-01	OH84B	RESULTS OF HEAT TRANSFER TEST IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER-VON KARMAN FACILITY TUNNELS A AND B UTILIZING SPACE SHUTTLE ORBITER THIN SKIN THERMOCOUPLE MODELS 56-O, 60-O, AND 83-O TESTS: OH84B, OH 105, IH-102
2464	160,829	V-02	OH84B	RESULTS OF HEAT TRANSFER TEST IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER-VON KARMAN FACILITY TUNNELS A AND B UTILIZING SPACE SHUTTLE ORBITER THIN SKIN THERMOCOUPLE MODELS 56-O, 60-O, AND 83-O TESTS: OH84B, OH 105, IH-102
2464	160,830	V-03	OH84B	RESULTS OF HEAT TRANSFER TEST IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER-VON KARMAN FACILITY TUNNELS A AND B UTILIZING SPACE SHUTTLE ORBITER THIN SKIN THERMOCOUPLE MODELS 56-O, 60-O, AND 83-O TESTS: OH84B, OH 105, IH-102
2464	160,831	V-04	OH84B	RESULTS OF HEAT TRANSFER TEST IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER-VON KARMAN FACILITY TUNNELS A AND B UTILIZING SPACE SHUTTLE ORBITER THIN SKIN THERMOCOUPLE MODELS 56-O, 60-O, AND 83-O TESTS: OH84B, OH 105, IH-102
2464	160,832	V-05	OH105	RESULTS OF HEAT TRANSFER TEST IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER-VON KARMAN FACILITY TUNNELS A AND B UTILIZING SPACE SHUTTLE ORBITER THIN SKIN THERMOCOUPLE MODELS 56-O, 60-O, AND 83-O TESTS: OH84B, OH 105, IH-102
2464	160,833	V-06	IH102	RESULTS OF HEAT TRANSFER TEST IN THE ARNOLD ENGINEERING DEVELOPMENT CENTER-VON KARMAN FACILITY TUNNELS A AND B UTILIZING SPACE SHUTTLE ORBITER THIN SKIN THERMOCOUPLE MODELS 56-O, 60-O, AND 83-O TESTS: OH84B, OH 105, IH-102
2465	167,674		OS55/57	AERODYNAMIC VENTING CHARACTERISTICS TESTS OF FULL-SCALE SPACE SHUTTLE MODEL 81-O HRSI TPS TILES UNDER A SIMULATED LAUNCH ENVIRONMENT IN THE NAS /ARC 9X7-FOOT WIND TUNNEL (OS55/57)

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DMS-DR REPORT NUMBER	NASA CR NUMBER		NASA SERIES NUMBER	SPACE SHUTTLE VEHICLE WIND TUNNEL TEST DATA REPORT TITLE
2466	167,663	V-01	OA257	RESULTS OF INVESTIGATIONS OF THE 0.010-SCALE OV-102 CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 72-0 IN THE NASA/LANGLEY RESEARCH CENTER 20-INCH MACH 6 TUNNEL (OA257)
2466	167,664	V-02	OA257	RESULTS OF INVESTIGATIONS OF THE 0.010-SCALE OV-102 CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 72-0 IN THE NASA/LANGLEY RESEARCH CENTER 20-INCH MACH 6 TUNNEL (OA257)
2467	160,834		IH103	RESULTS OF AEROTHERMODYNAMIC HEAT TRANSFER TESTS ON 0.0175-SCALE MODELS 60-OT AND 56-0/60T CONDUCTED IN THE NASA/AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL (IH103)
2468	167,352		OH105B/OH84C	RESULTS OF A HEAT TRANSFER TEST SERIES IN THE NASA/ARC 3.5 FOOT HYPERSONIC WIND TUNNEL UTILIZING SPACE SHUTTLE ORBITER THIN-SKIN THERMOCOUPLE MODELS 60-0 AND 83-0(TESTS OH84C AND OH105B)
2469	167,367		OS302A	SPACE SHUTTLE AFRSI LARGE-SCALE DEVELOPMENT TEST USING MODEL 117-0 SPECIMENS AND MODEL 96-0 TEST FIXTURE IN THE AMES RESEARCH CENTER 11X11-FOOT TRANSONIC WIND TUNNEL (OS302A)
2470	167,658		OS31A	SPACE SHUTTLE LRSI THIN TILE TEST IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT UNITARY PLAN WIND TUNNEL USING TEST FIXTURE 96-0 (OS31A)
2471	160,514		LA132	RESULTS OF TESTS ON A .02 SCALE SPACE SHUTTLE LAUNCH VEHICLE MODEL (89OTS) IN THE LARC 16-FT TRANSONIC WIND TUNNEL TO DETERMINE PRESSURE DISTRIBUTION ALONG THE EXTERNAL TANK LOX CABLE TRAY (LA132)
2472	160,494		OH400	RESULTS OF AN ORBITER SILTS POD HEAT TRANSFER AND FLOW FIELD TEST USING A 0.0175-SCALE SPACE SHUTTLE ORBITER(92-0) IN THE AEDC VKF HYPERSONIC WIND TUNNEL B (OH400)
2473	167,388	V-01	OA252	AERODYNAMIC LOADS TEST OF 0.66-SCALE SPACE SHUTTLE ORBITER TILE ARRAY MODEL (106-0) IN THE NASA/ARC 2-FOOT TRANSONIC WIND TUNNEL (OA252)
2473	167,389	V-02	OA252	AERODYNAMIC LOADS TEST OF 0.66-SCALE SPACE SHUTTLE ORBITER TILE ARRAY MODEL (106-0) IN THE NASA/ARC 2-FOOT TRANSONIC WIND TUNNEL (OA252)
2474	160,826		FA28	RESULTS OF TESTS ON A .004 SCALE SPACE SHUTTLE LAUNCH CONFIGURATION (MODEL 74-OTS) IN THE NASA/MSFC 14-INCH TRISONIC WIND TUNNEL (FA28)
2475	160,509		LA140	PRESSURE DISTRIBUTION AND INTEGRATED LOADS AT FOUR STATIONS ON THE SPACE SHUTTLE TANK LOX FEEDLINE (LA140)

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2476			IA190A/IA190B	** DOCUMENTATION NOT COMPLETE **
2477	160,825		LA141A/B	RESULTS OF INVESTIGATIONS ON AN 0.004-SCALE 140C MODIFIED CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL (74-0) IN THE NASA/LANGLEY RESEARCH CENTER 20-INCH MACH 6 TUNNEL (LA141)
2478	160,503	V-01	LA131	HIGH SUPERSONIC RUDDER EFFECTIVENESS AND EFFECT OF SILTS POD ON A 0.20-SCALE (REMOTELY DRIVEN CONTROL SURFACE) MODEL 106-0 SPACE SHUTTLE ORBITER TESTED IN THE NASA/LARC 4-FOOT UNITARY PLAN WIND TUNNEL (LA131)
2478	160,504	V-02	LA131	HIGH SUPERSONIC RUDDER EFFECTIVENESS AND EFFECT OF SILTS POD ON A 0.20-SCALE (REMOTELY DRIVEN CONTROL SURFACE) MODEL 106-0 SPACE SHUTTLE ORBITER TESTED IN THE NASA/LARC 4-FOOT UNITARY PLAN WIND TUNNEL (LA131)
2478	160,505	V-03	LA131	HIGH SUPERSONIC RUDDER EFFECTIVENESS AND EFFECT OF SILTS POD ON A 0.20-SCALE (REMOTELY DRIVEN CONTROL SURFACE) MODEL 106-0 SPACE SHUTTLE ORBITER TESTED IN THE NASA/LARC 4-FOOT UNITARY PLAN WIND TUNNEL (LA131)
2479			IA600	** DOCUMENTATION NOT COMPLETE **
2480	167,657		IH104	RESULTS OF HEAT TRANSFER TESTS ON THE SPACE SHUTTLE SECOND STAGE ASCENT VEHICLE AT FREESTREAM MACH=5.3 AND 7.3 IN THE NASA/ARC 3.5-FOOT HWT USING THE 0.0175-SCALE MODEL 60-OT(IH104)
2481	167,377		IA602	RESULTS OF TESTS IN THE NASA/MSFC 14-INCH TRISONIC WIND TUNNEL ON A 0.004-SCALE MODEL (74-OTS) THRUST AUGMENTED SPACE SHUTTLE INTEGRATEDVEHICLE (IA602)
2482	160,814	V-01	OA400	RESULTS OF TESTS FOR FORCE, MOMENT, PRESSURE AND AEROELASTIC DATA USING THE 0.030 SCALE PRESSURE LOADS SPACE SHUTTLE ORBITER MODEL (47-0) IN THE NASA/ARC 11 FOOT UNITARY PLAN WIND TUNNEL. (OA400)
2482	160,815	V-02	OA400	RESULTS OF TESTS FOR FORCE, MOMENT, PRESSURE AND AEROELASTIC DATA USING THE 0.030 SCALE PRESSURE LOADS SPACE SHUTTLE ORBITER MODEL (47-0(IN THE NASA/ARC 11 FOOT UNITARY PLAN WIND TUNNEL. (OA400)
2482	160,816	V-03	OA400	RESULTS OF TESTS FOR FORCE, MOMENT, PRESSURE AND AEROELASTIC DATA USING THE 0.030 SCALE PRESSURE LOADS SPACE SHUTTLE ORBITER MODEL (47-0(IN THE NASA/ARC 11 FOOT UNITARY PLAN WIND TUNNEL. (OA400)
2483	167,357	V-01	OS49	RESULTS OF A TEST OF THE FULL-SCALE NASA ORBITER VERTICAL TAIL (MODEL 111-0) IN THE AEDC 16 FOOT PROPULSION WIND TUNNEL (OS49)

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2483	167,358	V-02	OS49	RESULTS OF A TEST OF THE FULL-SCALE NASA ORBITER VERTICAL TAIL (MODEL 111-0) IN THE AEDC 16 FOOT PROPULS23[%2[& 1[[]6 .3\49.
2484			LA144	** DOCUMENTATION NOT COMPLETE **
2485	167,361		OS50/OS50A	RESULTS OF VENT PORT TPS LOADS TESTS IN THE AMES RESEARCH CENTER (ARC) 11X11-FOOT WIND TUNNEL USING MODEL 113-0 (OS50/OS50A)
2486	167,368	V-01	OA253	RESULTS OF WIND TUNNEL TEST OA253 IN THE AEDC 16-T PROPULSION WIND TUNNEL USING A 0.035-SCALE SS LAUNCH VEHICLE MODEL 84-OTS & ENTRY VEHICLE MODEL 84-0
2486	167,369	V-02	OA253	RESULTS OF WIND TUNNEL TEST OA253 IN THE AEDC 16-T PROPULSION WIND TUNNEL USING A 0.035-SCALE SS LAUNCH VEHICLE MODEL 84-OTS & ENTRY VEHICLE MODEL 84-0
2487	167,362		OS43/OS51/OS51B/OS	RESULTS OF AMES GAP FILLER TESTS USING TEST FIXTURE 96-0 IN THE NASA/AMES 11X11-FOOT TUNNEL (OS43,OS51,OS51B,OS51C)
2488	160,835		OS300	PRELIMINARY SCREENING TESTS OF THE SPACE SHUTTLE AFRSI MATERIAL USING MODEL 115-0 IN THE NASA/AMES RESEARCH CENTER 2X2 FOOT TRANSONIC WIND TUNNEL (OS300)
2489	167,366		OS56	RESULTS OF A WIND TUNNEL TEST ON THE SPACE SHUTTLE UMBILICAL PURGE CURTAIN IN THE AEDC 16-T PROPULSION WIND TUNNEL (PWT), USING MODEL 108-0 (OS56)
2490	167,349	V-01	OH109	TEST RESULTS FROM THE NASA/ROCKWELL INTERNATIONAL SPACE SHUTTLE 0.0175-SCALE ORBITER MODELS 56-0/60-0 AND 0.04-SCALE ORBITER FOREBODY MODEL 83-0 CONDUCTED IN THE AEDC/VKF-B 50-INCH HYPERSONIC WIND TUNNEL (TESTS OH109 & OH109B)
2490	167,350	V-02	OH109	TEST RESULTS FROM THE NASA/ROCKWELL INTERNATIONAL SPACE SHUTTLE 0.0175-SCALE ORBITER MODELS 56-0/60-0 AND 0.04-SCALE ORBITER FOREBODY MODEL 83-0 CONDUCTED IN THE AEDC/VKF-B 50-INCH HYPERSONIC WIND TUNNEL (TESTS OH109 & OH109B)
2490	167,351	V-03	OH109	TEST RESULTS FROM THE NASA/ROCKWELL INTERNATIONAL SPACE SHUTTLE 0.0175-SCALE ORBITER MODELS 56-0/60-0 AND 0.04-SCALE ORBITER FOREBODY MODEL 83-0 CONDUCTED IN THE AEDC/VKF-B 50-INCH HYPERSONIC WIND TUNNEL (TESTS OH109 & OH109B)
2491	167,659	V-01	OA258	RESULTS OF INVESTIGATIONS ON THE 0.020-SCALE OV-102 CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 106-0 IN THE USAF/AEDC VKF TUNNEL B (OA258)
2491	167,660	V-02	OA258	RESULTS OF INVESTIGATIONS ON THE 0.020-SCALE OV-102 CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 106-0 IN THE USAF/AEDC VKF TUNNEL B (OA258)

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2491	167,661	V-03	OA258	RESULTS OF INVESTIGATIONS ON THE 0.020-SCALE OV-102 CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 106-0 IN THE USAF/AEDC VKF TUNNEL B (OA258)
2491	167,662	V-04	OA258	RESULTS OF INVESTIGATIONS ON THE 0.020-SCALE OV-102 CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 106-0 IN THE USAF/AEDC VKF TUNNEL B (OA258)
2492	167,359		OH107	RESULTS OF THE SSV ELEVON GAP HEATING TESTS USING THE 0.025-SCALE SPACE SHUTTLE ORBITER MODEL (94-0) IN THE AEDC/VKF HYPERSONIC WIND TUNNEL B (OH107)
2493	167,665	V-01	OA259	RESULTS OF INVESTIGATIONS OF THE 0.010-SCALE OV-102 CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 72-0 IN THE NASA/AEDC VKF TUNNEL B (OA259)
2493	167,666	V-02	OA259	RESULTS OF INVESTIGATIONS OF THE 0.010-SCALE OV-102 CONFIGURATION SPACE SHUTTLE VEHICLE ORBITER MODEL 72-0 IN THE NASA/AEDC VKF TUNNEL B
2494	167,360		OH108	AERODYNAMIC HEATING TESTS OF A 0.10-SCALE SS ORBITER ELEVON/ELEVON GAP MODEL 93-0 IN THE NASA/ARC 3.5 FOOT HYPERSONIC WIND TUNNEL (OH108)
2495	160,844		OH110	TEST RESULTS FROM THE NASA/ROCKWELL INTERNATIONAL SPACE SHUTTLE 0.0175-SCALE ORBITER MODELS 56-0/60-0 AND THE 0.04-SCALE ORBITER FOREBODY MODEL 83-0 CONDUCTED IN THE NASA/ARC 3.5-FOOT HYPERSONIC WIND TUNNEL (TEST OH110)
2496	167,380	V-01	OH111	RESULTS OF THE TRANSATLANTIC ABORT MANEUVER TEST(OH111) USING THE 0.0175-SCALE 56-0 AND 60-0, AND THE 0.04-SCALE 83-0 THIN SKIN THERMOCOUPLE MODELS IN THE AEDC VKF TUNNEL B HYPERSONIC WIND TUNNEL(OH111)
2496	167,381	V-02	OH111	RESULTS OF THE TRANSATLANTIC ABORT MANEUVER TEST(OH111) USING THE 0.0175-SCALE 56-0 AND 60-0, AND THE 0.04-SCALE 83-0 THIN SKIN THERMOCOUPLE MODELS IN THE AEDC VKF TUNNEL B HYPERSONIC WIND TUNNEL(OH111)
2496	167,382	V-03	OH111	RESULTS OF THE TRANSATLANTIC ABORT MANEUVER TEST(OH111) USING THE 0.0175-SCALE 56-0 AND 60-0, AND THE 0.04-SCALE 83-0 THIN SKIN THERMOCOUPLE MODELS IN THE AEDC VKF TUNNEL B HYPERSONIC WIND TUNNEL(OH111)
2497			MA34	** DOCUMENTATION NOT COMPLETE **
2498	167,656		OA255/OA256	RESULTS OF SPACE SHUTTLE ORBITER (MODEL 70-0) LATE ENTRY RCS YAW JET EFFECTS TESTS IN THE NASA/LARC UPWT AND 16-FT. WIND TUNNELS (OA255/OA256)
2499	160,836		OA164	RESULTS OF TESTS USING A 0.36-SCALE MODEL (76-0) OF THE SSV ORBITER 101 IN THE NASA/AMES RESEARCH CENTER 40X80-FOOT SUBSONIC WIND TUNNEL(OA164)

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2500	160,848	OS301	PHASE II SCREENING TEST OF AFRSI MATERIAL USING MODEL 115-0 IN THE AMES RESEARCH CENTER 2X2-FOOT TRANSONIC WIND TUNNEL (OS301)
2501	167,373	OS304A	SPACE SHUTTLE AFRSI OMS PODS/JOINTS DEVELOPMENT TEST USING MODEL 116-0 SPECIMENS & MODEL 96-0 TEST FIXTURE IN THE AMES RESEARCH CENTER 11X11-FOOT TRANSONIC WIND TUNNEL (OS304A)
2502	167,378	OS304B	SPACE SHUTTLE AFRSI OMS PODS/JOINTS DEVELOPMENT TEST USING MODEL 116-0 SPECIMENS AND MODEL 81-0 TEST FIXTURE IN THE AMES RESEARCH CENTER9X7-FOOT SUPERSONIC WIND TUNNEL (OS304B)
2503	167,363	OS53A/OS53B	RESULTS OF COMBINED LOADS ORBITER TEST (CLOT) IN THE NASA/LARC 8-FOOT TPT USING THREE CONFIGURATION 20 TPS FLOW TEST PANELS (OS53A/B)
2504	167,379	OS302B	SPACE SHUTTLE AFRSI LARGE-SCALE DEVELOPMENT TEST USING MODEL 117-0 SPECIMENS AND MODEL 81-0 TEST FIXTURE IN THE AMES RESEARCH CENTER 9X7-FOOT SUPERSONIC WIND TUNNEL (OS302B)
2505	167,376	OS46A-G	RESULTS OF ASCENT AERODYNAMIC LOADING TESTS OF THE SS THERMAL PROTECTION SYSTEM (TPS) IN & AROUND THE ORBITER/ET UMBILICAL DOOR & CAVITY, USING MODELS 108-0 & 1090 IN THE AEDC 16-T PROPULSION WIND TUNNEL (OS46A-G)
2506	167,384	OS60,1,2,3	GAP FILLER REUSE TESTS OF FULL-SCALE SPACE SHUTTLE ORBITER TILE ARRAY MODELS IN THE NASA/ARC 9X7-FOOT AND 11-FOOT UNITARY PLAN WIND TUNNEL (OS60,OS61A,OS61B,OS62,OS62A, AND OS63)
2507	167,683	MA33A/B	RESULTS OF INVESTIGATIONS OF THE SPACE SHUTTLE ORBITER ONE-QUARTER-HERTZ OSCILLATION ANOMALY IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT AND 9X7-FOOT WIND TUNNELS USING 0.02-SCALE MODEL 106-0 (MA33A/B)
2508	167,650	OS306A/B	SPACE SHUTTLE AFRSI DESIGN CRITERIA DEVELOPMENT TESTS IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT AND 9X7-FOOT WIND TUNNELS USING MODEL 23-0 (OS306A/B)
2509	167,654	OA307A/B	SPACE SHUTTLE FRCI-12 TPS TILE VENTING TEST IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT AND 9X7-FOOT WIND TUNNELS (OA37A/B)
2510	167,651	OS309A	SPACE SHUTTLE AFRSI FULL-SCALE CREDIBILITY TEST IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT WIND TUNNEL USING MODEL 124-0 INSTALLED IN THE 96-0 TEST FIXTURE (OS309A)

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2511	167,669	V-01	IA300	RESULTS OF COLD PLUME TESTS OF THE 0.010-SCALE MODEL (75-OTS) IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT WIND TUNNEL (IA300)
2511	167,670	V-02	IA300	RESULTS OF COLD PLUME TESTS OF THE 0.010-SCALE MODEL (75-OTS) IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT WIND TUNNEL (IA300)
2511	167,671	V-03	IA300	RESULTS OF COLD PLUME TESTS OF THE 0.010-SCALE MODEL (75-OTS) IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT WIND TUNNEL (IA300)
2512	167,667		OA308	BOUNDRY LAYER TESTS OF THE SPACE SHUTTLE AFRSI MATERIAL IN THE NASA/AMES RESEARCH CENTER 2X2-FOOT TRANSONIC WIND TUNNEL (OA308)
2513	167,678		OS313	SPACE SHUTTLE AFRSI GAP FIX TEST OS313 IN THE AEDC/USAF 16T TRANSONIC PROPULSION WIND TUNNEL USING MODEL 129-0 INSTALLED IN THE MODEL 96-0 TEST FIXTURE
2514	167,687		FA301	** DOCUMENTATION NOT COMPLETE **
2515	167,684		OS305-1/5	POST-TEST DATA REPORT FOR THE SPACE SHUTTLE FULL-SCALE AFRSI SEQUENCE OF ENVIRONMENTS TEST (OS305-1 TO 5) IN THE NASA/AMES RESEARCH CENTER 11X11-FOOT WIND TUNNEL
2516	167,688		OS311	** DOCUMENTATION NOT COMPLETE **
2517	167,689		OS314A/B/C	** DOCUMENTATION NOT COMPLETE **