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	MARS EXPLORATION VENUS SWINGBY AND CONJUNCTION CLASS MISSION MODES TIME PERIOD 2000 TO 2045
	By Archie C. Young, John A. Mulqueen, and James E. Skinner Program Development
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16. ABSTRACT

Trajectory and mission requirement data is presented for Earth-Mars opposition class and conjunction class round trip stopover mission opportunities available during the time period year 2000 to year 2045. The opposition class mission employs the gravitational field of Venus to accelerate the space vehicle on either the outbound or inbound leg. The gravitational field of Venus was used to reduce the propulsion requirement associated with the opposition class mission. Representative space vehicle systems are sized to compare the initial mass required in low Earth orbit of one mission opportunity with another mission opportunity. The interplanetary space vehicle is made up of the spacecraft and the space vehicle acceleration system. The space vehicle acceleration system consists of three propulsion stages. The first propulsion stage performs the Earth escape maneuver, the second stage brakes the spacecraft and Earth braking stage into the Mars elliptical orbit and effects the escape maneuver from the Mars elliptical orbit. The third propulsion stage brakes the mission module into an elliptical orbit at Earth return. The interplanetary space orbit.

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TECHNICAL MEMORANDUM

MARS EXPLORATION VENUS SWINGBY AND CONJUNCTION CLASS MISSION MODE TIME PERIOD 2000 TO 2045

INTRODUCTION

This report presents information on performance, operation requirements and their sensitivity for the Venus swingby and conjunction class stopover missions to Mars. The time period considered to develop this information is year 2000 to 2045. Initial mass required in Earth orbit was determined for each launch opportunity associated with the two classes of missions. The Venus swingby mode allows a short stopover time, 60 days, at Mars and can be considered as ε precursor mission for the long stopover time, up to 550 days, at Mars which occurs with the conjunction class missions.

A manned space station, to be built and serviced with the aid of the Space Shuttle, will open space to a limitless range of opportunities for explorations. Using the Space Station as a stepping stone, manned missions to Mars can be achieved during the 21st century. With the technology being developed today, the space industry could be mining large amounts of material, expanding our economic activities in space and bringing the benefits back to Earth. The data contained in this report can be used for NASA planning, research and advanced technology programs.

Direct trajectories to and from Mars for a roundtrip stopover mission, stay time up to 60 days, require very high propulsive energy at some of the departure and/or arrival planets. This type of mission is referred to as opposition class mission. A method used for reducing this energy requirement is the Venus swingby mode. A Venus swingby utilizes the gravitational field of Venus to either accelerate or decelerate the space vehicle as it passes by the planet. An accelerated effect is desired for an outbound Venus swingby enroute from Earth to Mars and a decelerated effect is desired for an inbound Venus swingby enroute from Mars to Earth. The conjunction class mission requires relatively low propulsive energy for all the powered maneuvers effected at the planets. This low energy requirement can be achieved by optimizing the stopover time at Mars so that near-Hohmann type transfers can be made on both the outbound and inbound legs.

ASSUMPTIONS

The pertinent assumptions used in this study to develop the Mars mission opportunities and resultant requirements are given for the departure and capture orbit parameters, propulsion stage and planetary spacecraft elements. Figure 1 gives the assumptions for the Venus swingby mission mode and Figure 16 gives the assumptions for the conjunction class missions. The interplanetary space vehicle was assumed to be assembled in and depart from the 270 n.mi. altitude Space Station circular orbit. Mars capture and escape orbit was considered to be an elliptical orbit with a 24-hr period and periapsis altitude of 270 n.mi. The interplanetary mission module returns to an Earth elliptical orbit with a 24-hr period and a perigee altitude of 270 n.mi. The interplanetary spacecraft required velocity increments are achieved by three propulsion stages. The first propulsion stage effects the Earth escape maneuver, the second stage brakes the spacecraft and Earth braking stage into the Mars elliptical capture orbit and effects the escape maneuver from the Mars elliptical orbit. The third propulsion stage brakes the mission module into a 24-hr elliptical orbit at Earth return. Each of the three propulsion stages has a mass fraction (λ) of 0.90 and uses cryogenic LOX/LH₂ propellant with a specific impulse of 480 sec. The mass fraction of 0.90 is relatively low for the propulsive stage loading required for the Mars mission, but this low mass fraction was assumed in order to account for micro-meteoroid protection and propellant boiloff. Propellant boiloff was not considered to be a problem since insulation technology would result in insulations that would solve this problem.

Venus swingby, outbound, inbound, or double swingby, was used to lower the energy required for the Mars opposition class missions. The Venus closest approach distance was constrained to be equal to or greater than 0.1 planet radii (330 n.mi.).

For the conjunction class missions, type I (<180 deg) or type II (>180 deg) transfer trajectories were considered. The Mars stopover time was optimized to achieve minimum initial weight in Earth orbit.

The weight of the interplanetary spacecraft for the Venus swingby mission is as follows. Weight of the mission module is 83,000 lb and the probes weight is 42,000 lb. The Mars excursion module weight equals 95,000 lb. The interplanetary spacecraft weight for the conjunction class mission is as follows. Weight of the mission module is 117,000 lb, the Mars excursion module weighs the same as for the Venus swingby mission mode, and the pioneer Mars base weight is 57,000 lb.

MISSION OPPORTUNITIES

Mission opportunities for standard direct flights to Mars and return will occur near the Earth-Mars opposition, and precede by 90 to 180 days the opposition dates which occur on the average every 26 months. Two general classes of direct roundtrip mission modes to Mars are available; they are (1) opposition class mode with short stopover time at Mars and (2) conjunction class mission mode with long stopover time at Mars.

Because of the eccentricity of Mars orbit, the mission trajectory profile changes from one opposition to the next. The cyclic pattern of mission profile variation repeats every 15 years or every 7 oppositions [1]. The relative positions of the Earth-Mars oppositions are indicated in Figure 2 for two periodic cycles of oppositions from year 2000 to 2031. The slight inclination of the Mars orbit with respect to the ecliptic plane causes an interplanetary transfer trajectory also to be inclined to the ecliptic, but this effect is small compared to the effect caused by the eccentricity. The relative position of Earth and Mars for an opposition class mission, stopover time at Mars approximately 60 days, causes the energy requirement to be excessive for the propulsive stages to perform a standard roundtrip Mars mission. One approach to reduce the required energy level for the opposition class mission is to use the gravity field of Venus either enroute to Mars for an outbound swingby or enroute to Earth for an inbound swingby. The cause of high energy required for a standard roundtrip mission is due to the following relative planetary motion. The flight time for a near-Hohmann outbound leg is such that, at Mars arrival, Earth is ahead of Mars in heliocentric longitude, i.e., Mars arrival occurs after opposition. This makes it impossible to employ a near-Hohmann transfer for the inbound leg; the required heliocentric transit angle must greatly exceed the Hohmann transfer angle of 180 deg. Thus, it is never possible to leave Earth on a minimum energy transfer outbound leg and arrive at Mars soon enough to leave Mars on a minimum energy inbound leg. The relative position of Earth at Mars arrival can be adjusted with a swingby of Venus enroute to Mars on an outbound leg or a swingby of Venus enroute to Earth on an inbound leg. The major advantage of making a swingby of Venus is that the hyperbolic encounter with the planet changes the velocity of the space vehicle relative to the Sun. The magnitude of the velocity change can be large enough to make a significant desirable change in the heliocentric trajectory. The high energy level required can be avoided in the conjunction class mission mode where near-Hohmann transfers can be used on both the outbound and inbound leg by adjusting the stay time at Mars appropriately.

The availability of a Venus swingby mode can be determined by the following facts [1]: (1) The space vehicle passes inside or near the orbit of Venus either on the outbound leg or on the inbound leg of a direct roundtrip mission to Mars. Figure 3 illustrates these conditions for an outbound leg and an inbound leg. (2) The gravity field of Venus is sufficiently powerful to significantly shape the interplanetary transfer trajectory in a desirable way. (3) The angular rate of Venus orbit is large compared to that of Mars, so that Venus is generally available either on the outbound leg or on the inbound leg.

The initial step in determining a Venus swingby trajectory profile for a given mission opportunity is the determination of the relative heliocentric position of the three planets, Venus, Earth, and Mars. A plot of the relative position of the three planets was developed to determine whether an inbound or an outbound Venus swingby was available near a given Earth-Mars opposition. A procedure which has proven useful is a continuous plot of the dates of conjunction (Earth-Venus Alignment), alignment of Venus-Mars, and opposition (Earth-Mars alignment) [4]. Figures 4, 5, and 6 give the plots of the planets continuous relative position for years 2000 to 2031. Considering the first outbound Venus swingby, a search is made for a region in which the Earth-Venus conjunction is followed, approximately 200 days later, by a Venus-Mars alignment which, in turn, is followed 200 days later by an Earth-Mars opposition. Looking at Figure 4, it can be predicted that an inbound swingby associated with the 2001 opposition would prove desirable because of the proper relative position of the planets alignments. For the 2003 opposition, the time phasing between appropriate planetary position is almost equal for the outbound and inbound swingby, but successive events are separated by something less than 200 days. Thus, Venus may be available both for the outbound and inbound leg. However, the usefulness of the double swingby must be determined by detailed trajectory analysis. The symmetrical alignment of the planets about an opposition occurs every third opposition. Based on an analysis of the planetary positions, Venus swingby should be available as shown in Figures 4, 5, and 6. Having narrowed the area down where an opportunity exists, the problem is to locate the precise launch and arrival dates which will allow an economic transfer to Venus, followed by an economic transfer to Mars.

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MISSION CHARACTERISTICS

Mission characteristic data has been developed for launch opportunities during the years 2000 to 2031 for opposition class missions via Venus swingby and for launch opportunities during the years 2030 to 2045 for conjunction class missions.

Figure 7 presents the Venus swingby, outbound or inbound, employed for each of the opposition class missions during the time period. Also given in Figure 7 is the year and month of the launch window for Earth departure and the total mission time associated with each opportunity. The shortest total trip time experienced is 558 days for the September 2007 launch opportunity employing an inbound Venus swingby; this launch opportunity precedes the December 2007 Earth-Mars opposition. The longest total trip time required during the time period considered is 737 days for the March 2028 launch opportunity employing a double Venus swingby, i.e., swingby Venus on both the outbound and inbound leg.

Table 1 contains the mission characteristics for the Venus swingby profiles. Information is given for the Julian date (J.D.) and the speed, hyperbolic excess speed given in units of Earth mean orbital speed (EMOS), at each one of the planets. The Earth-Mars opposition date is also given. The Earth launch window date precedes the Earth-Mars opposition date from 3 to 6 months for the inbound Venus swingby mission mode. For the outbound Venus swingby mission mode, the Earth launch window occurs 12 to 19 months before the Earth-Mars opposition. The minimum value for hyperbolic excess speeds at Earth and Mars is approximately 0.10 EMOS for near-Hohmann type transfer trajectories. From Table 1, it can be seen that an economical propulsion energy requirement would be associated with the 2001 inbound swingby opportunity since the speeds on the outbound leg are near a minimum value at Earth departure and Mars arrival. The speed at Mars departure, however, is slightly high with a value of 0.2050 EMOS. The Earth arrival speed of 0.1418 EMOS is relatively low for an opposition class mission.

Mars conjunction class stopover mission summary data is given in Figure 19. The summary data contains dates of Earth-Mars opposition, Earth launch dates, stopover times at Mars, and total mission time. The minimum total mission time is 950 days for the April 2033 launch opportunity; the optimum stopover time at Mars is 550 days for this opportunity. The maximum total mission time is 1004 days for the June 2035 launch opportunity; the optimum stopover time at Mars for this opportunity is 530 days.

Table 22 gives the mission characteristics for the opportunities during the time period year 2030 to 2045. The Earth launch date precedes the Earth-Mars opposition by three to four months which can be seen by comparing the Earth-Mars opposition dates with the leave Earth dates. Most of the hyperbolic excess speeds at the planets are close to a minimum value of approximately 0.10 EMOS. The shortest trip time occurring during this time period is 200 days for the outbound and inbound legs of the year 2033 opportunity. The longest trip time is 356 days for the outbound leg of the year 2037 opportunity. The data contained in this table should cover the envelope of values for the conjunction class stopover mission since it covers one complete synodic period of Earth and Mars.

INTERPLANETARY SPACE VEHICLE

The interplanetary space vehicle is made up of the spacecraft and the space acceleration system. The interplanetary space vehicle was assumed to be assembled in and depart from the Space Station circular orbit.

The spacecraft acceleration system is made up of three propulsion stages. The first propulsion stage effects the Earth escare maneuver, the second stage brakes the spacecraft and Earth braking stage into the Mars elliptical capture orbit and effects the escape maneuver from the Mars elliptical orbit. The third propulsion stage brakes the mission module into a 24-hr elliptical orbit at Earth return.

The total weight required in the Space Station orbit for the interplanetary space vehicle has been determined for the Mars mission employing the Venus swingby mission mode and the conjunction class mission. The weight in the Space Station orbit can be compared from one opposition opportunity to another for the two classes of Mars missions considered.

The time period considered for the Venus swingby mission mode is years 2000 to 2031. The conjunction class mission mode launch opportunity considered is during the time period year 2030 to 2045. The Venus swingby mode has a stopover time at Mars of 60 days and could be used as a precursor mission for the conjunction class missions which have stopover times at Mars of up to 550 days.

The spacecraft is made up of a mission module (the living and work area for the crew), a Mars excursion module, and experimenter accommodations. A number of unmanned probes and orbiters are included to complement the manned activity. Major elements of the spacecraft are interconnected by pressurized tunnels allowing shirt-sleeve passage between them. A minimum crew of 6 is necessary to operate the space systems and perform a reasonable scientific exploration program.

The mission module is the control center for the entire space vehicle and provides a habitable living, operations, and experiment center for the mission crew. The basic mission module provides the environmental control, power system radiators, and the communications system for the most stringent mission. Provisions are made for incremental loading of meteoroid shielding, expendables, and system spares as necessary. The mission module contains all the subsystems necessary for life support, command and control functions, experiments analysis, and information transfer during the course of the mission. It is pressurized to a 7-psia (48.23(10³) Newton/m²) oxygen-nitrogen atmosphere, providing a viable shirtsleeve environment for the crew.

The Mars excursion module transports three of the crew members and equipment from the space vehicle in Mars orbit to the Mars surface. It provides living quarters and a laboratory during the 30-day stay on the Mars surface; then transports the crew, scientific data, and samples back to the orbiting vehicle.

Two to five days are spent surveying the planet for landing sites, performing orbital experiments (including deployment of probes), and preparing the Mars excursion module for operation. Three of the six-man crew then descend to the planet surface in the Mars excursion module. After aeroballistic entry, the Mars excursion module is slowed by a ballute retardation system, and, using propulsion descends to the surface. After a 30-day stay on the planet, the ascent module of the Mars excursion module brings the three men and scientific payload back to the space vehicle. During planetary operations, the men in the space vehicle continue the

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orbital experimentation, monitor the planetary operations. and maintain the space vehicle operations. The ascent vehicle is discarded in ι planet orbit after the crew has transferred to the mission module [5]. Figure 8 shows the spacecraft used in the Venus swingby mission mode.

Interplanetary trajectory parameters (launch dates, trip times, heliocentric transfer angles, etc.) have been determined which result in a minimum total initial weight to be assembled in the Space Station's orbit. The three variable size propulsion stages were sized using general scaling weight laws which are dependent upon propellant loading and whose coefficients are input to the interplanetary trajectory shaping program. Up to five major interplanetary maneuvers can be optimized.

The propulsion stage weights for each opposition class mission opportunity during year 2600 to 2031 are given in Figures 9 and 10. The stage weight required for the first propulsion stage ranges from 870,000 to 2,720,000 lb. Propulsion stage weight required for the second stage ranges from 353,000 to 1,094,000 lb, and the Earth braking propulsion stage, stage three, weight ranges from 24,000 to 102,000 lb.

The initial mass required in low Earth orbit for each mission opportunity is given in Figures 11 and 12. The initial mass required ranges from 1,470,000 to 4,065,000 lb. The initial mass in Earth orbit can be equated to cost and used to determine the favorable mission opportunities. Double swingbys are available every third pportunity and are generally characterized by having a higher energy requirement han the other opportunities. However, there is an exception for the opportunity in year 2007 where the energy requirement is greater than the double swingby year 2010. Generally there should be two favorable opportunities for every unfavorable, high-energy requirement opportunity.

The interplanetary space vehicle configuration is given in Figure 13 for the opposition class mission via Venus swingby for the year 2025 opposition opportunity. The spacecraft length and information for each of the propulsion stages is given. The interplanetary space vehicle total length is 276.5 ft. Also given is the propellant loading for each of the propulsion stages and total initial weight required in the Space Station orbit. The total initial weight required for this mission opportunity is 2.08 million pounds.

Propulsion stage and interplanetary weight data for the Mars conjunction class stopover mission is given in Figures 21 and 22. The propulsion stage weights for each conjunction class mission during year 2030 to 2045 are <u>riven</u> in Figure 21. The stage weight for the first propulsion stage ranges from 600,000 to 725,000 lb. Propulsion stage weight required for the second stage ranges from 112,000 to 169,000 lb. The Earth braking stage, stage three, weight ranges from 25,000 to 36,500 lb. The initial mass required in low Earth orbit for each mission opportunity is given in Figure 22. The mass requirement ranges from 1,010,000 to 1,170,000 lb.

The interplanetary space vehicle sized to perform the conjunction class mission for the year 2031 opposition opportunity is given in Figure 23. The total vehicle length is 211.5 ft for the three propulsion stages plus the spacecraft. The propellant loadings are given for each of the propulsion stages. The total weight required in low Earth orbit is 1.17 million pounds.

INTERPLANETARY TRAJECTORY CALCULATIONS

The computer program used in this work to compute the interplanetary trajectory characteristics is based on the restricted two-body (patched conic) approximation of the interplanetary space vehicle trajectory. While the vehicle is within the sphere of influence of Venus, the swingby planet, it is assumed to be on a free-flight hyperbolic trajectory about Venus, and gravitational effects of all other bodies are neglected. There is no change of energy with respect to the swingby planet, Venus. Conservation of energy requires that the magnitude of the vehicle's velocity, relative to Venus, as it leaves the sphere of influence of Venus must equal to the magnitude of its velocity as it enters the sphere of influence approaching Venus. If the required angle of deflection, bend angle, at Venus is too large to be achieved by constraining the periapsis altitude to one-tenth of the planet radii, a propulsive maneuver is effected in conjunction with the Venus gravity field to give the required bend angle.

Because of the relative positions of Earth, Venus and Mars, a compromise transfer profile has to be made for the opposition class mission mode, over a true optimum Hohmann transfer on the outbound and inbound legs. The pertinent trajectory parameters for each of the launch opportunities during year 2000 and 2031 are given in Tables 2 through 18. Description of each of the parameters is given in the Appendix. The hyperbolic excess velocity at each of the planets is defined in terms of C_3 , right ascension and declination. The heliocentric trajectory inclination relative to the ecliptic plane, semi-major axis, and eccentricity are also given in the

Independent optimization of each leg is possible when the conjunction class roundtrip missions are considered. The outbound leg takes place near one opposition, and by adjusting the stopover time at Mars appropriately, the inbound leg will take place near the following opposition. Examination of single leg trajectory data [2] indicates that if the outbound and inbound legs of a roundtrip mission could be optimized separately, then departure and arrival hyperbolic excess speeds at both Earth and Mars of less than 0.10 to 0.15 EMOS (Earth Mean Orbital Speed of 97,700 ft/sec) could be attained. The total mission time for conjunction class missions is greater than the mission time of the Venus swingby opposition class mission; 950 to 1004 days, for conjunction class compared to 558 to 737 days for Venus swingby. Tables 23 through 29 give the trajectory data for the conjunction class missions for opportunities occurring during the years 2030 through 2045.

REPRESENTATIVE MISSION PROFILES

Representative mission profiles are given for Venus swingby and conjunction class mission modes. The Venus swingby is used for the opposition class missions with short stopover times at Mars to reduce the total propulsive energy required to perform the mission. The conjunction class mission mode can optimize the stopover time at Mars in order to employ minimum energy trajectories for both the outbound and inbound legs.

Figure 14 illustrates mission profiles for the inbound Venus swingby associated with the year 2014 opposition opportunity and outbound Venus swingby for the year 2025 opposition opportunity. Both Venus swingby missions have a 60-day stopover time at Mars. On the inbound Venus swingby mission profile, the heliocentric transit angle of the Earth to Mars outbound leg is slightly greater than 180 deg; because of

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the higher average angular rate of Earth orbit, Earth is about 45 deg ahead of Mars at Mars arrival. This angular position of Earth relative to Mars necessitates a large transfer angle on the inbound leg with perihelion occurring inside Venus orbit in order to attain a sufficiently large angular rate to catch up with Earth. Utilizing the swingby of Venus on the inbound leg permits a trajectory with the high angular rate needed to overtake Earth and results in near-tangential trajectory conditions for departure from Mars and arrival at Earth. The near-tangential mission profile associated with a Venus swingby reduces the propulsive energy required compared to a standard Mars opposition class mission. The total mission time for the year 2014 opposition opportunity is 634 days. Decailed characteristics for this mission are given in Tables 1 and 9.

The year 2025 outbound Venus swingby has essentially the same desirable characteristics as the favorable inbound swingby. The outbound swingby is characterized by a heliocentric transfer angle between Earth and Venus of over 180 deg; with a transfer angle between Venus and Mars of less than 180 deg. The total angle is slightly over 360 deg. Of paramount importance is the fact that the average angular rate of the outbound trajectory is much greater than that of Earth in its orbit. Thus, Earth is far behind Mars at Mars arrival, i.e., Mars arrival occurs much sooner than opposition. This situation permits, as shown, a near-Hohmann Mars-Earth trajectory to be utilized on the inbound leg. The total mission time for the year 2025 outbound Venus swingby opposition opportunity is 614 days. Detailed characteristics for this mission are given in Tables 1 and 14. The characteristics of the outbound and inbound Venus swingby trajectories will yield lower required initial weight in low Earth orbit to perform the mission.

The mission profile for the conjunction class mission associated with the year 2031 opposition opportunity is given in Figure 24. Earth launch date occurs on December 25, 2030 and the outbound trajectory to Mars has a trip time of 282 days; arriving at Mars on October 3, 2031. Stopover time at Mars, for year 2031 opposition opportunity is 500 days; this length of stopover time allows for a near-Hohmann type trajectory on the return leg to Earth. Mars departure date for the Earth return leg is February 14, 2033 and the return trip time to Earth is 216 days; arriving back at Earth September 18, 2033. The total mission time for the year 2031 opposition opportunity is 998 days. Minimum energy trajectories were used both on the outbound and inbound leg. Detailed characteristics of this mission are given in Tables 22 and 23.

CONCLUSION

Optimum transfers for opposition class missions to Mars via Venus swingby have been computed for the attractive launch and arrival dates between years 2000 and 2031. Also optimum transfer for conjunction class missions to Mars have been completed for the attractive opportunities between years 2030 and 2045.

It is possible to employ an outbound or inbound Venus swingby for every Earth-Mars opposition; oppositions occur approximately every 26 months. Venus swingby permits the heliocentric transfer trajectory to be nearly tangential relative to Earth and Mars orbit upon planet departure and arrival; thus reducing the required propulsive maneuver energy requirement. Two out of every three mission opportunities to Mars should be favorable when employing the Venus swingby mode. The mission time is increased from 20 to 50 percent employing the Venus swingby mode over the direct flights to Mars.

Summary results obtained for the Venus swingby mission are presented in Figure 15. The initial mass required in low Earth orbit over the 31-year time period ranges from 1.47 to 4.1 million pounds. If the unfavorable opportunities are disregarded, the initial mass required would not exceed 3.0 million pounds. Stage weight required for the Earth escape stage, first stage, ranges from 870,000 to 2,720,000 lb. The first stage weight is approximately 2-1/2 times the weight of the second stage. Mission duration for the Venus swingby mode ranges from 558 to 737 days.

The short stopover time, approximately 60 days, at Mars associated with the opposition class mission may be extended to over 100 days by utilizing Venus swingby on both the outbound and inbound leg during the favorable opportunities. It may also be possible to launch one vehicle to Mars via Venus swingby and have it arrive at Mars just prior to launching a second vehicle which proceeds directly to Mars. The two vehicles could rendezvous at Mars and still have time for an economical return to Earth utilizing a Venus swingby. Other combinations are feasible which utilize this unique flexibility.

Optimum roundtrip trajectories for the conjunction class missions to Mars and return can be achieved by adjusting the stopover time at Mars. Near-Hohmann type trajectories can be employed both on the outbound and inbound leg with the conjunction class mission. Data has been developed for one Earth-Mars synodic period between years 2030 and 2045 which consists of seven launch opportunities associated with the oppositions occurring during this time period. The minimum and maximum envelop requirement for any other opportunity will be represented by the data covered during this synodic period.

Summary results of work completed on the conjunction class mission is presented in Figure 25. Mission opportunity for the conjunction class mission occurs approximately every 26 months. Initial mass required in low Earth orbit ranges from 1.01 to 1.17 million pounds. The escape stage, the first stage, weight minimum and maximum is 600,000 and 725,000 lb, respectively. The first stage weight is about four times the weight of the Mars braking and escape stage, the second stage. The variation in total mission time ranges from 950 to 1004 days. Stopover time at Mars varies from 340 to 550 days; the shorter stopover time is associated with longer outbound and inbound leg flight time. Earth departure energy, C₃, varies from 8.9 to 16.2 km²/sec² across the 15 years considered.

Requirements for the conjunction class mission can be extrapolated for opportunities occurring before or after the time period years 2030 to 2045 since the requirements are approximately repetitive every seven opportunities. For example, the requirements for the year 2029 conjunction class opportunity should be similar to the year 2044 opportunity requirements. Figures 2 and 17 give information when opportunities occur before or after the year 2030 to 2045 time period.

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TIME PERIOD OF CONSIDERATION: YEAR 2000 TO 2031

PLANET DEPARTURE AND CAPTURE ORBIT PARAMETERS

CIRCULAR ORBIT ALTITUDE = 270 N. MI 24 HR ELLIPTIC ORBIT PERIAPSIS ALTITUDE = 270 N. MI 24 HR ELLIPTIC ORBIT PERIAPSIS ALTITUDE = 270 N. MI 24 HR ELLIPTIC ORBIT PERIAPSIS ALTITUDE = 270 N. MI
EARTH DEPARTURE MARS CAPTURE MARS ESCAPE EARTH CAPTURE

HELIOCENTRIC PROFILE

VENUS MINIMUM CLOSET APPROACH EQUAL 0.1 PLANET RADII (330 N. MI) VENUS SWINGRY MODE (OUTBOUND, INBOUND OR DOUBLE SWINGBY)

INTERPLANETARY SPACE VEHICLE

SPACECRAFT:	MISSION MODUI	LE WEIGHT	= 83,900 LBS
	MARS EXECURS	HON MODULE WEIGHT	= 95,000 LBS
	PROBES WEIGHT	F	= 42,000 LBS
PROPULSION STAGES	FIRST STAGE	SECOND STAGE	THIRD STAGE
MASS FRACTION (λ)	0.90	0.90	0.90
ISP (SEC)	480	480	480
PROPELLANT	LOX/LH2	LOX/LH2	LOX/LH2

Figure 1. Study assumptions for Venus swingby mode.



Figure 2. Earth-Mars opposition year 2000 to 2031 time period.

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Direct stopover mission to Mars, example profile for opposition class stopover mission. Figure 3.

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	2007	Q	YES		•	•	500	
	POSITION 2005	YES	Q			•		- 000
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AILABILITY	ΥЕА 1999	YES	ON	JIBUTIUN ABONINAS ONTOBNI	•	•	50 •	3 LIAN DATE
AV	SION	TBOUND SWINGBY	OUND SWINGBY	H-VENUS ALIGNMENT S-MARS ALIGNMENT H-MARS MENT (OPPOSITION)	•	•	2001	- 70 5000
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Figure 4. Planetary position data year 1999 to 2007.

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	INBOUND SWINGB	>	MAYBE	ON	YES	MAYBE	Q	YES		
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AVAILABILITY OF SWINGBYS

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Figure 5. Planetary position data year 2010 to 2020.

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		Σ	NOISSII			2020	YEAR* OF 2022	: EARTH- 2025	MARS OPP 2027	OSITION 2029	2031	
		0	UTBOUND	SWINGBY		Q	MAYBE	YES	Q	MAYBE	YES	
		5	NBOUND S	WINGBY		YES	MAYBE	Q	YES	MAYBE	ON	
PLANET		 0+●	O EARTI O EARTI O EARTI (OPPOC	H-VENUS A S-MARS AL H-MARS AI SITION)	LIGNME IGNMEN			- <i>N</i> ₁ ,	JISVIIVA ABSNINS ONTOBO	OUTBOUND SWINGBY SWINGBY AVAILABL		
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AVAILABILITY OF SWINGBYS

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Figure 6. Planetary position data year 2020 to 2031.

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STOPOVER TIME EQUAL 60 DAYS TIME PERIOD 2000 TO 2031

NOISSIM	EARTH LAUNCH DATE	TOTAL TRIP TIME (DAYS)
INBOUND SWINGBY	MARCH 2001	909
OUTBOUND SWINGBY	AUGUST 2002	610
OUTBOUND SWINGBY	JUNE 2004	659
INBOUND SWINGBY	SEPTEMBER 2007	558
DOUBLE SWINGBY	JANUARY 2009	736
OUTBOUND SWINGBY	NOVEMBER 2010	650
INBOUND SWINGBY	NOVEMBER 2013	634
INBOUND SWINGBY	NOVEMBEE, 2015	577
OUTBOUND SWINGBY	APRIL 2017	638
INBOUND SWINGBY	JUNE 2020	594
OUTBOUND SWINGBY	OCTOBER 2021	636
OUTBOUND SWINGBY	SEPTEMBER 2023	614
INBOUND SWINGBY	NOVEMBER 2026	570
DOUBLE SWINGBY	MARCH 2028	737
OUTBOUND SWINGBY	JANUARY 2030	654

Figure 7. Mars stopover mission with Venus swingby.

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Interplanetary spececraft for Mars opposition class mission. Figure 8.

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Figure 13. Interplanetary vehicle sized for 2025 opposition Venus outbound swingby opportunity.

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TWO MISSION OPPORTUNITIES OUT OF EVERY THREE SHOULD BE FAVORABLE

INITIAL MASS REQUIRED IN LOW EARTH ORBIT RANGES FROM 1.47 TO 4.1 MILLION POUNDS

WEIGHT RANGE OF PROPULSION STAGES

THIRD STAGE: 24,000 TO 102,000 POUNDS

SECOND STAGE: 353,000 TO 1,094,000 POUNDS

FIRST STAGE: 870,000 TO 2,720,000 POUNDS

• MISSION TIME RANGES FROM 558 TO 737 DAYS

● EARTH DEPARTURE C₃ RANGE FROM 9.2 TO 37.0 KM²/SEC²

Figure 15. Venus swingby summary results.

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TIME PERIOD OF CONSIDERATION: YEAR 2030 TO 2045

PLANET DEPARTURE AND CAPTURE ORBIT PARAMETERS

H DEPARTURE CIRC Capture 24 H Escape 24 H Capture 24 H

HELIOCENTRIC PROFILE

MARS STOPOVER TIME OPTIMIZED TO MINIMIZE INITIAL WEIGHT IN EARTH ORBIT TYPE I OR TYPE II TRANSFER TRAJECTORY

INTERPLANETARY SPACE VEHICLE

MISSION MODULE WEIGHT = 117,000 LBS Mars execursion module weight = 95,000 LBS Pioneer Mars Base = 57,000 LBS
SPACECRAFT: MIS MA PIO

Figure 16. Study assumptions for conjunction class missions.



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Figure 17. Earth-Mars opposition for years 2030 to 2061.

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Direct stopover mission to Mars, example profile for conjunction class stopover missions. Figure 18.

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DATE OF OPPO	SITION	EARTH LAUNCH	H DATE M	ARS STOPOVER TIME	TOTAL MISSION TIME
МАҮ	2031	DECEMBER	2030	(DAYS) 500	(DAYS) 998
JUNE	2033	APRIL	2033	550	950
SEPTEMBER	2035	JUNE	2035	530	1004
NOVEMBER	2037	AUGUST	2037	340	986
JANUARY	2040	SEPTEMBER	2039	340	18 8
FEBRUARY	2042	OCTOBER	2041	340	066
MARCH	2044	NOVEMBER	2043	340	966

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Figure 19. Mars conjunction class stopover mission.

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Figure 20. Interplanetary spacecraft for Mars conjunction class missions.

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Stage weight as a function of year of Earth-Mars opposition. Figure 21.

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Interplanetary vehicle sized for 2031 opposition opportunity for conjunction class mission. Figure 23.

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Mars conjunction class mission profile for 2031 opposition. Figure 24.

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MISSION OPPORTUNITY OCCURS APPROXIMATELY EVERY 26 MONTHS

INITIAL MASS REQUIRED IN LOW EARTH ORBIT RANGES FROM 1.01 TO 1.17 MILLION POUNDS

WEIGHT RANGE OF PROPULSION STAGE

 THIRD STAGE:
 25,000 TO 36,500 POUNDS

 SECOND STAGE:
 112,000 TO 169,000 POUNDS

 FIRST STAGE:
 600,000 TO 725,000 POUNDS

STOPOVER TIME AT MARS RANGES FROM 340 TO 550 DAYS

TOTAL MISSION TIME RANGES FROM 950 TO 1004 DAYS

EARTH DEPARTURE C₃ RANGES 8.9 TO 16.2 Km²/SEC²

Figure 25. Conjunction class summary results.

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TABLE 1. MARS STOPOVER MISSION WITH VENUS SWINGBY MISSION DEFINITION

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MISSION	EARTH- MARS OPPOSITION J.D. 2450000 OR 2460000	L V EARTH J.D. 2450000 OR 2460000 (EMOS)	PASS VENUS	ARRIVE MARS	LEAVE MARS	PASS VENUS	ARRIVE EARTH	TOTAL MISSION TIME (DAYS)
2001 INBOUND SWING: Y	2073	1996 (.1018)		2182 (.1283)	2242 (.2050)	2436 (.2531)	2602 (.1418)	606
2003 OUTBOUND SWINGBY	2880	2510 (.1272)	262: (.1781)	2812 (.2183)	2872 (.1623)		3120 (.2343)	610
2005 OUTBOUND SWINGBY	3683	3164 (.1549)	3330 (.3259)	3505 (.2061)	3565 (.1230)		3823 (.1256)	6 59
2007 INBOUND SWINGBY	4460	4350 (.1741)		4532 (.2028)	4592 (.2446)	4747 (.3308)	4908 (.1515)	558
2010 DOUBLE SWINGBY	5225	4854 (.1735)	4983 (.2463)	5196 (.1456)	5256 (.2294)	5433 (.2370)	5590 (.1497)	736
2012 OUTBOUND SWINGBY	5991	5528 (.1778)	5689 (.3570)	5842 (.1795)	5902 (.1402)		6178 (.2535)	650
2014 INBOUND SWINGBY	6755	6618 (.1162)	1	6888 (.1229)	6948 (.2135)	7083 (.3321)	7252 (.1491)	634
2016 INBOUND SWINGBY	7529	7336 (.1957)		7596 (.1642)	7656 (.2447)	7834 (.3958)	7913 (.1828)	577
2018 OUTBOUND SWINGBY	8325	7854 (.1709)	8007 (.3345)	8198 (.1865)	8258 (.1223)		8492 (.1886)	638
2020 INBOUND SWINGBY	9134	9010 (.1612)		9196 (.1326)	9256 (.2114)	9437 (.2764)	9604 (.1365)	594
2022 OUTBOUND SWINGBY	9922	95.8 (.1229)	9666 (.2012)	9820 (.1946)	9880 (.1736)		60154 (.2744)	636
2025 OUTBGUND SWINGBY	0694	0196 (.1679)	0358 (.3758)	0494 (.2146)	0554 (.1216)		0810 (.0947)	614
2027 INBOUND SWINGBY	1455	1346 (.1349)		1554 (.1978)	1614 (.2132)	1751 (.3440)	1916 (.1614)	570
2029 DOUBLE SWINGBY	2219	1856 (.1711)	2008 (.2460)	2210 (.1789)	2270 (.1864)	2437 (.2374)	2552 (.1956)	737
2631 OUTBOUND SWINGBY	2989	2530 (.1674)	2693 (.3516)	2884 (.1712)	2944 (.1265)		3184 (.2764)	654

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MISSION MARS STOPCVER MISSION WITH INBOUND VENUS SWINGBY DATA FOR 2001 OPPOSITION ¢, TABLE

3.90 56.83 17.84 3.87 56.83 17.84 3.94 56.83 17.84 3.85 55.83 17.84 3.84 56.83 17.84 3.83 56.83 17.84 5 3.82 .1 56.83 4 17.84 044 03-40 04-80 04-80 05-80 ETA0 8.91892 5.46773 1.79050 1.03424 19.22890 8.85718 5.46331 1.79050 1.03424 19.31134 18.35405 1.03424 18.74673 8.75360 5.45553 1.79858 1.83424 18.68886 18.64826 18.6473 ---- --- 2.17 779 93.4 351.0 10.5 --- 11308 -6.80 1333 85.2 164.9-39.1 .624 --- 3.57 .886 90.7 332.8 13.4 1 .62A side trailing edge .00000 --- --- 2.77 .780 93.3 349.2 13.6 --- 11368 -6.80 1.353 85.2 164.9-39.1 .624 --- 3.57 .886 90.7 332.8 13.4 DARK SIDE TRAILING EDGE .00000 ---- 3.03 .780 33.3 348.4 14.9 --- 1.398 -6.80 1.353 85.2 164.9-39.1 .624 --- 886 90.7 332.813.4 DARK SIDE TRAILING EDGE ---- 2.35 780 93.4 350.4 11.4 --- 2.80 1.353 85.2 164.9-39.1 .624 --- 3.57 886 90.7 33281 13.4 DARK SIDE TRAILING EDGE .00000 HARS UENUS EARTH ---- 2.55 .780 93.4 349.9 12.5 --- 2.80 1353 85.2 164.9-39.1 .624 --- 3.57 .886 90.7 332.8 13.4 DARK SIDE TRAILING EDGE .00000 DECL 347.5 16.3 164.9-39. 332.8 13.4 ---- 2:00 779 93.4 351.4 9.5 --- 1.398 -6.80 1.353 85.2 164.9-39. -64 --- 3:57 185 90.7 332.8 13.4 Dark side 1railing Edge .0000 ETA4 1.03424 • • • 1.03424 1.03424 PLANET PLANET PLANET ar ar ar 3.31 .780 93.3 3 -6.80 1.353 85.2 3.57 .886 90.7 3 E TRAILING EDGE ເບເບເ DEPARTURE PASSAGE ARRIVAL 154 154 ETA3 1.79850 8.77714 5.45739 1.79850 1.79958 1.79050 പ I Z U Z I Z U Z I Z U Z CONDITION ETAR 5.45992 5.45421 5.45333 --- --- 1.398 --- 1.398 --- 1.398 --- 1.398 DARK SIDE PERIH APHEL PERIH APHEL PERIH APHEL ETAl 8.81072 8.73842 8.73048 9 1.193 12.4 163.3 9 1.193 12.4 163.3 6 1866 286.1 534.1 5 217.77 .00 125.23 9 1.193 14.3 163.2 7 1.054 176.8 340.3 5 .806 286.1 537.7 5 217.77 .00 125.23 7 1.193 16.2 163.1 7 1.054 276.8 340.3 8 .806 286.1 537.7 5 217.77 .00 125.23 1 1.192 20.1 153.0 7 1.054 176.8 340.3 6 .806 286.1 537.7 5 217.77 .00 125.23 THET1 THET2 F THET1 THET2 F THET1 THET2 F THET1 THET2 F AS DECS ETA 1 22.1 163.0 1 176.8 340.3 2 26.1 537.7 .00 125.23 1.193 10.6 163.4 1.054 176.8 340.3 .806 286.1 537.7 217.77 .00 125.23 1.192 18.2 163.6 1.654 176.8 340.3 .806 286.1 537.7 217.77 .00 125.23 GAM4 1.3A499 1.30099 1.30099 1.30099 1.30099 1.30899 1.30099 RAS CARD 1.98987 1.98987 1.98987 1.98987 1.98987 1.98987 1.191 1.054 1.054 217.77 1.98987 ECCEN SHA ECCEN SHA ECCEN SHA ECCEN SHA 87.4 .178 88.4 .327 101.6 .226 81.08-37.55 2 -4.01 1.083 88.5 .168 6.29 .695 88.4 .327 -.98 1.640 181.6 116.8 46.7 P1.08-37.55 51986.0 52182.0 9.71 240.5 -64.4 -3.76 1.082 88.2 169 52242.0 52435.6 27.30 351.8 20.3 6.29 .695 88.4 .327 52435.6 52602.0 58.29 210.4 -4.6 -.98 1.240 101.6 .226 1.106 53.6 .4141 .908 2.218 116.8 46.7 81.08-37.55 1.8658 1.2539 3.73870 1.23852 2.14643 1.41676 -62.0 -3.54 1.082 87.9 .169 20.3 6.29 .695 88.4 .327 -4.6 -.98 1.240 101.6 .226 2.218 116.8 46.7 81.08-37.55 -3.35 1.081 87.7 169 6.29 .695 88.4 .327 -.98 1.240 101.6 .226 116.5 46.7 81.08-37.55 -55.2 -3.04 1.080 87.1 .170 20.3 6.29 581.4 .327 -4.6 -981.240 101.6 2.218 116.8 46.7 81.08-37.55 -2.91 1.079 86.8 .171 6.29 .695 88.4 .327 -.98 1.240 101.6 .226 116.8 45.7 81.08-37.55 2 E 1.42378 7E114.1 56494.3 1.40735 1.46228 1.40089 L ISU ISU ISU ISU 8 -3.19 1.081 6.29 .695 -.98 1.240 1 116.8 46.7 2.15597 2.13309 1.23852 2.13990 2.13586 2.13405 2.13591 ANTAX INH GAMI 1.23852 1.23852 1.23852 1.23852 -53.1 26.3 -4.6 2.218 EH ECC ž -67.0 20.3 2.4.6 2.218 52182.0 9.19 238.2 524 52435.6 37.30 351.8 52682.0 58.29 210.4 53.6 4141 9988 2 53.6 1.2539 1.26413 9.10 237.2 -37.30 351.8 58.29 210.4 4141 .998 2 1.2411 150 3.23870 9.16 236.2 -37.36 351.8 58.29 216.4 4141 .998 8 1.298 8 1.298 8 1.23870 241.6 351.8 210.4 51988.6 52182.6 9.39 239.3 52242.6 52435.6 37.36 351.8 52435.6 52662.6 58.29 216.4 1.166 53.6 .4141 .968 1.255986 1.62183 3.23870 3.58086 1.62183 3.23870 6 37.38 351.8 6 37.38 351.8 9 58.29 218.4 10.18 24 37.30 35 58.29 21 MARS VENUS EARTH . т 53.6 .4141 1.2539 1.66303 3 51992.0 52182.0 9.10 52242.0 52435.6 37.30 52435.6 52602.0 58.29 1.106 53.6 .4141 3.56783 1.59850 3 53.6 .4141 1.2539 1.59143 . . . 51994.0 52182.0 52242.0 52435.6 52435.6 52602.0 1.1066 53.6 .4 51996.8 52182.6 52242.6 52435.6 52435.6 52682.6 1.186 53.6 4 1.253 3.57196 1.6884 51984.0 52182.0 52242.0 52435.6 52435.6 52602.0 1.106 53.6 .4 3.61607 1.6530 PLANET PLANET PLANET 51996.0 52182.0 52242.0 52435.6 52435.6 52602.0 1.106.53.6 STOP PASS ARRIUE KAPPA DEPARTURE Passage Arrival 3.56798 .57207 CORD1 DU1 LAUNCH DEPART PASS RCP

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MARS STOPOVER MISSION WITH OUTBOUND VENUS SWINGBY MISSION DATA FOR 2003 OPPOSITION с. TABLE

C3-PD C3-AD DUCP ---- --- -6.91 1.278 88.6 99.6-59.7 28.13 719 ---- -3.83 654 95.2 333.4-24 42.26 LIGHT SIDE LEADING EDGE .833 1.401 -2.01 1.045 103.6 214.0-22.9 48.69

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 -- -3.89
 .694
 94.7
 332.9-25.0
 40.99

 2.02
 1.360
 89.4
 1.00.3
 -- -- -3.89
 .694
 94.7
 332.9-25.0
 40.99

 128.4
 98.2
 244.3
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 LIGHT SIDE
 LEADING EDGE
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 .00072

 128.4
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 441.9
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 441.9
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 1.03.7
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TABLE 4. MARS STOPOVER MISSION WITH OUTBOUND VENUS SWINGBY MISSION DATA FOR 2005 OPPOSITION

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	_	C3-P C3-P	C3-R	ETA0	99.5 38.9 13.6	6166.	38.4 38.4 13.9	.1188	97.1 38.6 13.9	.7189	94.0 38.3 13.6	1.5314	33.93 13.6	1.5424	8.46 7.76	1.4583	5.E1	4.547
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		HETI	HEL	_	73.1 51.9 83.1	1.2	52.52 85.46	1.2	51.1	1.2	179.2 49.6 83.1	1.2	40.0 40.0 10.0	1.2	182.1 58.9 185.4	1.2	183.8 58.6 185.6 185.4	1.2
		SHA T SHA T SHA T	SHA 1	GANG	- 789 1 - 789 1 - 1.62 - 1.173 1	1.36764	.789 1.027 212.63 1.173	1.37250	211.629 211.629	1.37250	1.173 269.59 1.173	1.36754	200-22 200-59 200-59	1.36764	202- 1.031 1.173	1.37250	2100.1 210.61 1.031 1.173	1.37250
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		I IS I IS I AP	1 15	e S	87.2 76.6 98.7	2.03	87.6 76.5 313.65- 91.2	2.02	88.6 76.8 313.25 91.2	2.01	89.7 77.2 311.98 90.7	2. 0 1	90.4 90.7 90.75 90.75	2.81	90.8 77.1 912.96 91.2	2.0 0	91.5 77.1 913.80 91.2	2.06
à				Ę	840 345 771	42035		41536	9.9 272	39747	.346 .346 .6.8	.38274	845 546 771	38356	841 546 772	.38931		.39593
		MMA MAAN X	-	3	499 - 199 -	ณ์ ด	9.40 9.61 9.22 2.22	N.	9.93 9.93 9.55 9.55 9.55	ณ์	8.4.38 7.6.4 4.1.6	N.	8.4 6.4 7.6 7.6 7.6 7.6 7.6 7.6 7.6 7.6 7.6 7.6	a) a	0.0 0.0 0.0 0.0 0.0 0.0 0.0 0.0 0.0 0.0	n D	20 - 20 - 20 - 20 - 20 - 20 - 20 - 20 -	2
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		αα <u>α</u> €€	€ œ	C0RD3	162.5 53.8 528 328.0	47368	162.3 54.6 527 325.9	456964 466964	159.5 54.4 542 325.9	45635	155.4 53.4 582 328.0	47368	153.9 53.4 328.6	47368	153.6 54.6 3256 3256	+ F F 6 6 4 .	152.1 54.2 3256 325.9	40004.
	• EARTH	64-53 64-53	C3-RS	202	22.70 98.59 98.59 13.23	816 1.	22.47 99.02 4071 13.43	2019 201 1.	21.65 95.72 95.72 13.43	979 820 1.	20.97 92.03 13970 13.23	369 468 1.	21.01 92.03 92.03 92.03	3099 468 1.	21.28 92.51 .3968	530 1.	21.58 92.51 3961 13.43	979 536 1.
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WITH INBOUND VENUS SWINGBY MISSION MARS S'FOPOVER MISSION WITH INBOUND V DATA FOR 2007 OPPOSITION ч С TABLE

	PERIH AF Perih Af Perih Af	ETAI	-989	16.31778	.991 - .568 - DARK	16.4578B	.993 - 568 - DARk	6.53755		6.63330		6.74121		DARK
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	THET1 THET1 THET1 THET1 RAS DE	ENA	9 336.2 9 205.0 2 69.1 2 69.1	18 1.	8 337.7 8 206.1 269.1	79 1.	9 339.0 2 206.1 2 69.1	1 6	2696.1	9 1.0	341.8 265.1 269.1 269.1	9	243.1	
	2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	Ū	1.23	2.369	1.461 1.246 .78 .78	2,442	1.246	2.4427	1.478 1.246 320.72	2.4427	1.482 1.246 783 320.72	2.4427	1.246	320.72
	9000 9000 9000 9000	Щ2 М2		929E		11151	-7.18	2405	.327 .448 .248 -7.18	3749	.328 .448 .274 .7.18	5159	.329 448 475	-7.18
	I ISA I ISA I Sd I Sd I ISA	GA	95.8 107.5 105.4 213.12	5 1.9	95.4 108.2 212.98	2 1.9	95.1 108.2 108.2 212.98	• 1.9	94.8 108.2 212.98	1.9	94.5 168.2 165.4 212.98	9.1	94.2 108.2	212.98
	1111 1000	GAM1	1.140 .627 .214	2.6312	1.140	2.5944	1.141	2.5704	1.143 .631 1.214 12.4	2.54916	1.231	2.53001	1.145	12.4 2.51316
	I I I I I I I I I I I I I I I I I I I I	*	а. 7. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1.	4687	3.67 3.40 199.61	4687	3.40 1.36 1.09.00	1687	3.28 3.36 3.36 109.0	1687	3.12 3.36 3.36 109.0	1687	88.9 9,98 9,98 9,98 9,00	109.0
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"25"	A A A HO	DU3	99.79 9321.0 1321.0	1.05997	101.0 321.0 321.0	86293.	101.2 99.3 321.0 122.1 101.2 101.2	86695.	8.001 6.156 6.156 6.156 4.156 4.156	.20398	321.00.3 321.00.3 321.00.3	86595.	99.8 99.3 92.3	1555.1
	20-00-00 20-01-01 20-00-000-000-0000000000		32.10 58.50 58.50 54.4 54.4 50 50 50 50 50 50 50 50 50 50 50 50 50	138	30.51 53.05 98.95	961	29.45 53.69 98.99 3659 536 535	• 8Ee	28.51 53.09 98.99 4659 536	• • IE	27.66 53.69 98.99	• 620	26.98 53.09 98.59	4659
PLANET PLANET PLANET	STOP PASS ARIUE APPA	DU2	4530.0 4746.9 4988.0 37.7	3.11	4532.0 4746.9 4908.0 38.0	. e	1532.6 1746.9 1968.6 18.6	3.68	1532.8 1746.9 1908.8 18.0	Э.11:	1746.9 1746.9 18.8 8		532.0 968.8	8.0 1.65 3.182
PARTURE SSAGE PICAL	INCH BART CORDI	101	46.6.6 596.6 46.9 745.9 747.9 747.9 747.9 747.9 747.9 747.9 747.9 747.9 747.9	1.55375	342.6 5 96.9 5 46.9 5	. 48743	44.054 92.0554 46.954 1837 1837 1837 1837	.44373	46.9 54 46.9 54 46.9 54 1.07 1.08 1.08 1.08 1.08 1.08 1.08 1.08 1.08	. 40449	48.0 54 46.9 54 16.9 54	36923	90.00 90.00 90.00 90.00 90.00	.107 3 1.0882 1.33766
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MARS STOPOVER MISSION WITH DOUBLE VENUS SWINGBY MISSION DATA FOR OUTBOUND LEG FOR 2010 OPPOSITION ė TABLE

229.9 72.0 54.24 124.4 7.6 18.12 .00000 233.2 /3.0 53.99 127.1 8.5 18.57 .20000 54.14 4.86643 4.89688 53.21 18.43 53.54 17.83 53.80 18.80 4.82646 C3-PD C3-AD DUCP 6.07578 5.86198 52.79 18.45 6.13314 6.52987 ETA0 235.0 73.4 5 128.4 9.9 1 128.4 21559 231.5 72.6 5 126.0 7.6 1 .00000 238.7 73.8 5 128.6 15.7 1 1.87375 236.8 73.7 5 126.3 15.9 1 1.04667 240.7 73.9 128.4 17.5 1.34826 E ARTH UENUS MARS DECL . 09900 ETA4 00000 DEPARTURE PLANET • 1 PASSAGE PLANET • 1 ARRIVAL PLANET • 1 . 00000 • • . 00000 <u>a</u> a .726 ---- 10.75 1.261 91.5 .721 --- -2.83 .613 94.3 .Dark SIDE LEADING EDGE .728 --- 10.79 1.259 90.7 .716 --- -.87 .614 95.1 .Dark SIDE LEADING EDGE .727 --- 10.78 1.260 51.3 .720 --- -2.77 .616 95.5 Dark Side Leading Edge .728 --- 10.77 1.259 90.5 .715 --- -.78 .618 96.8 . Dark SIDE LEADING EDGE .727 --- 10.80 1.260 91.1 .719 --- -2.50 .618 96.3 Dark Side Leading Edge .727 --- 10.80 1.259 90.9 .718 --- -2.12 .620 97.1 .718 SIDE LEADING EDGE .728 ---- 10.74 1.258 90.3 .713 --- 1.39 .617 96.7 Dark SIDE LEADING EDGE ູ ISd ETAJ . 0000 2.44066 1.10314 .0000 . 00000 . 00688 . 69998 PERIM APHEL I 2 V 2 PERIM APHEL I 2 V 2 A PASS CONDITION 2.48265 i.10714 .00000 2.01741 1.06764 1.05678 ETA2 .88888 2.66310 1.12266 1.90237 ETAL .864 205.0 370.6 1.183 345.3 533.3 342.77 .00 125.76 54852.0 54981.5 25.84 24.8 -45.0 -8.08 933 93.1 154 .860 197.6 365.0 54981.5 55204.0 56.63 280.4 15.3 2.78 1.279 95.6 .393 1.180 340.2 532.0 1.100 63.3 .7082 .949 2.159 117.6 77.9 113.22 37.68 336.29 .00 125.33 .6461 .7082 1.3977 .13577 .13.68 336.29 .00 125.33 .6461 .7082 1.3977 .0000 2.48940 1.84999 .00000 .00000 . 54854.0 54982.5 26.72 23.9 -44.8 -8.15 .933 93.5 .155 .861 199.5 366.4 54982.5 55196.0 57.74 279.1 22.2 4.03 1.379 94.9 .393 1.183 342.4 528.8 1.100 56.5 .4124 .938 2.173 117.4 77.2 114.27 37.00 338.02 .00 125.23 .6541 .746E 1.3977 .00000 2.50915 1.57592 .000000 .00000 1 4.33013 2.14091 .00000 2.50915 1.57592 .000000 .00000 1 54983.5 27.64 23.0 -44.5 -8.22 .934 93.8 1.57592 .00000 .0000 55200.0 57.78 279.1 24.4 4.40 1.379 94.7 .392 1.183 343.4 530.0 54.5 .4118 .936 2.182 117.3 76.4 115.66 36.35 339.61 .00 125.44 7.7261 1.3977 THET1 THET2 THET1 THET2 RAS DECS ETA . 0000 CAM4 . 00000 GAMB ECCEN SHA ECCEN SHA DECP -43.7 -8.32 .935 94.5 .160 26.4 4.70 1.379 94.1 .398 2.245 116.5 74.7 118.01 34.61 Ê PSI 1 PSI 1 RAP ŝ I I U I I I U I ANNAX INH GAM1 4 DECL DECL 985.5 29.61 21.2 -210.0 56.86 279.4 2.9 686440 .941 2 .6865 1.3977 1.87214 .00000 **«** « Ŧ αœ Eng PLANET = EARTH PLANET = UENUS PLANET = MARS 04-63 04-63 54858.0 54984.5 2 54984.5 55204.0 5 1.152 53.2 .4082 .5701 .7082 CORD2 DU2 54856.0 54983.5 5 54983.5 55200.0 1 1.107 54.5 1 .6621 1.7261 .6621 1.2681 4.36836 1.99763 54848.0 54979.5 54979.5 55196.0 1.100 65.3 54850.0 54980.5 54986.5 55196.0 1.100 63.4 54860.0 54985.5 54985.5 55210.0 1.172 522.9 PASS ARRIVE RCP KAPPA CORD1 DV1 . 6782 . 6782 4. 45043 DEPARTURE P PASSAGE ARRIVAL DEPART PHSS RCP

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MISSIM MISSION WITH DOUBLE VENUS SWINGBY DATA FOR INBOUND LEG FOR 2010 OPPOSITION MARS STOPOVER 2 TABLE

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505 25 19.81 49.74 4.24855 4.43800 1 22.6 -0.52 0.648 107.2 0.424 1.261 207.0 353.6 ---- 2.35 1.398 88.1 261.8 13.8 51.13 7 70.0 9.82 1.255 91.5 0.151 0.855 348.5 543.7 0.726 0.965 -7.26 0.929 89.3 44.1-37.1 20.37 2.097 118.5 24.1 253.87-47.30 339.46 0.00 87.01 LIGHT SIDE TRAILING EDGE 0.00000 261.3 13.7 b1.61 14.1-37.1 25.37 0.0000 260.8 13.5 52.13 44.1-37.1 20.37 0.0000 £18 4-61584 4.80947 DEPARTURE PLANET - MARS PASSAGE PLANET - VENUS PASSAGE PLANET - VENUS ARTIVAL PLANET - EARTH THETI THET2 PERIH APHEL I 2 V 2 PSI 2 R A DECL RAS DECS ETA PASS CONDITION 261.7 14.6 43.2-36.6 49.00 0.727 0.987 -7.47 0.928 89.4 258.7 16.3 0.727 0.987 -7.47 0.928 89.7 43.4-37.4 Dates trailing (Date 0.0000 ETA4 1.82757 1.03784 0.00000 0.00000 -0.64 0.635 105.8 0.418 1.249 204.8 354.3 ---- 2.45 1.395 80.3 5.59 1.257 91.4 0.151 0.856 349.0 545.1 0.726 0.985 -7.09 9.929 80.1 117.8 83.9 253.55-47.10 340.64 0.00 88.02 LIGHT SIDE TRAILING EDGE 1.262 208.2 353.8 ---- 2.34 1.400 88.2 0.855 343.7 0.726 0.985 -7.26 0.929 89.3 339.46 0.60 87.47 LIGHT SIDE 19AILING EDGE 0.726 0.985 -7.26 0.929 89.3 LIGHT SIDE TRAILING EDGE ST3 1.03954 0.0000 1.94675 1.03784 0.0000 ETAR 1.83485 1.03715 ETAL 1.92538 **5525.0 55435.2 *8.43 87.8 21.5 -0.95 0.637 106.3 0.420** 1.255 205.7 357.9 **15435.3 55594.0 45.71 276.4 70.5 9.80** 1.257 31.2 **0.152 0.152 0.857 350.8 547.5 1.264 55.3 0.3322 1.096 2.156 117.6 83.2 249.68-45.54 344.21 0.00 93.17 1.5033 2.3754 1.3971 1.39783 0.00000 0.00000 1.39789 0.00000 1.39789 0.00000 1.39789 0.00000 1.39500 1.39789 0.00000 1.30500 1.30789 0.00000 1.30500 1.39780 0.00000 1.30500 1.39780 0.00000 1.30500 1.39780 0.00000 1.30500 1.39780 0.00000 1.30500 1.39780 0.00000 1.30500 1.39780 0.00000 1.30500 1.30780 0.00000 1.30500 1.39780 1.39780 0.00000 1.30780 0.00000 1.30780 1.39780 0.00000 1.30780 1.39780 0.00000 1.30780 1.39780 1.39780 0.00000 1.30780 1.397800 1.39780 1.397800 1.397800 1.3978000 1.3978000 1.3978000 1.3978000 1.3978000 1.39780000 1.39780000 1.395600** 1.276 209.3 354.1 0.855 348.5 543.7 330.46 0.00 87.95 ż 9.0000 0.0000 0.0000 0.0000 GAM3 0.0000 0.0000 9.000 ECCEN SHA ECCEN SHA DECP A 22.7 -0.51 9.544 107.9 0.427 70.0 9.32 1.256 91.5 0.151 2.091 118.6 83.9 253.20-47.32 -0.50 0.647 108.6 0.430 9.82 1.256 91.5 0.151 118.7 83.7 252.49-47.34 0.00000 2.47052 1.33146 Z 1.32569 0.000% 2.39470 1.33146 PSI 1 ISA 1 ISA 1 ISA 1 ISA 5 2.25405 T C T ïs 0.00000 22.3 ನನ ತಿಹಿತಿ Ž 22.8 78.9 2.885 55256.0 55433.5 46.68 87.9 2 55433.5 55590.0 44.27 276.7 2 1.271 55.6 0.3818 1.112 2. 1.4749 2.3764 1.3971 1.4749 2.3764 1.3971 3.82548 1.3674 0.00000 55260.0 55422.8 51.48 87.4 2 55432.8 55683.0 46.96 276.8 7 1.181 57.0 0.3932 1.077 2. 1.4654 2.3768 1.3971 2. 1.4654 2.3768 1.3971 4.11038 1.3475
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TABLE 8. MARS STOPOVER MISSION WITH OUTBOUND VENUS SWINGBY MISSION DATA FOR 2012 OPPOSITION

DEPARTURE PLANET Passage planet Arrival planet	- EARTH - UENUS - RARS													DEPARTI	14 14 14	ANET -	EARTH UENUS		
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RESTART RETURN CORDI COR	C3-RS I	R A DRD3	DECL	11	r 1	PSI 1	ECCEN	A SMR	THEL	T1 THE	ET2 PER	ILH APHE	- I 2	1 2 7	5 I S	Q.	DECL	UCP C3-RD	
DU1 DU2	20	(1	Put a	G	i M I	GA	۲ ۲	F	5	Gama	E	'A1	ETAZ	ũ	EA1	ETA	-	ETAO	
55520.0 55689.4 55689.4 55838.0 1170 31.8 51816 55176.0 1.4692 1.79 4.48616 2.796	30.48 31 117.50 21 4841 .4 16.75 7 16.75 7 16.75 7 1.3	34.0 24.0 24.0 24.0 24.0 24.0 24.0 24.0 2	6.7 1.6 650 10.4 2.723	3.49 29 105.9 -2.62 86 2	.840 1.373 5.5 .666 .5372	88.1 76.5 131.62 95.2	4 0 0 4 1 0	.759 1.158 29.85 1.286 1.45432	175.7 46.2 193.1	460.2 159.6 161.13 413.5		.989 RK 51DE 1.656	1.20 1.81 2.73 2.73	1.196 1 .633 1 .632 1 1.098 1 1.098 1	67.6 64.1 61.2	205.4 128.5 358.6	3.7 1 25.5 1 12.2	19.35 30.68 54.40	
55522.0 55689.4 55689.4 55842.0 1.187 31.8 5592.0 56178.0 1.4643 2.6796	29.52 35 115.71 23 4865 .4 17.44 7 17.44 7 1.2 1.2 1.2	33.2 5.9 717 3. 828 3.	6.3 11.1 2.823	3.46 .22 105.9 -2.57	.841 1.372 5.1 .664	88.9 76.7 131.22 95.6		.759 1.155 29.85 1.287	177.4 46.8 194.1	459.9 161.3 101.3 414.6		.938 .938 	1.67 1.84 2.75	1.196 1 1.196 1 01NG ED 1.099 1	000 000 01.0 01.0 01.0	. 99434 205.5 126.8 359.1	31. 26.6 69269 4.21	40664 17.52 28.87 57.00	
55524.0 55688.8 55688.8 55840.0 1.188 32.4 55900.0 56178 1.780 4.40197 2.718	28.45 33 112.56 23 4769 .4 16.79 7 18 1 2 1.76	ະ ເດີຍ ເດີຍ ເດີຍ ເດີຍ ເດີຍ ເດີຍ ເດີຍ ເດີຍ	5.3 5.83 5.83 1.1 2.7998	8 - 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5		98.8 77.8 138.87 95.2		.760 .760 1.159 28.86 1.285	179.9 44.9 193.1	4161.08	5532 5532 100 100	1 5.6 .987 RK SIDE 1.657	2.68 2.68 2.68 2.68	2.3755 1.197 1.632 1.632 1.632 1.699 1.099	8 1. 603.5 61.5 61.5	. 09937 204.8 127.6 354.2	36. 36. 25. 26. 26. 26. 26. 26. 26. 26. 26. 26. 26	80976 14.82 29.46 56.39	
55526.0 55688.8 55688.8 55840.0 1 1.201 32.3 4 5590.0 56178.0 4 1.4651 1.765 4.38865 2.7118	28.13 32 112.56 23 4752 .4 16.79 .7 1.76	629 9.7 89.6 3. 89.6 3. 829 3.	4.7 599 1.1 2.7998	3.40 1.06.1 -2.53 80 2.5 80 2.5 80 80 80 80 80 80 80 80 80 80 80 80 80	.844 1.373 .844 .9.5 .664	96.7 77.6 95.2 95.2	2 - 2 - 2 - 2 - 2 - 2 - 2 - 2 - 2 - 2 -	1.159 28.86 1.285	181.7 44.9 193.1	4450 1600.1 1482.00	-534 -534 -512 -512	8 5.5	2.68 1.82 2.68 2.68 1.82	2.3588 1.198 1.632	1 1. 003.5 01.4	69817 264.9 127.8 359.2	2000 2000 2000 2000 2000 2000 2000 200	36976 13.79 29.46 56.39	
55528.0 55688.8 55688.8 55842.0 1 1.211 32.2 .4 55902.0 56178.0 1.4661 2.6512 4.38601 2.6512	28.46 32 111.78 32 1734 .4 1734 .4 1.2 8 1.2 1.2 18 1.2 1	2553 3. 21:2 3. 21:2 3.	4.0 6.8 6.8 1.1 7.8 8231	3.30 .22 .6.1 -2.57	.845 1.373 3.0 1 .664	91.5 31.16 35.6 1.75	640 201 201 201 201 201 201 201		10.00 10	1001 1001 1001 1001 1001 1001 1001 100	.535 .535 .912 1.912	 24 51DE 1.662	2013 1.83 1.83 2.75 2.75	1 1 1 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	1 1. 667.5 61.5 61.5	69817 205.1 127.1 359.1	2005 2005 2005 2005 2005 2005 2005 2005	25581 28.59 28.59	
55530.0 55688.8 55688.8 55844.0 1 5504.2 55844.0 1 5504.2 56180.0 1.4661 2.5956 4.39435 2.5956	28.26 32 10.84 23 17.48 23 17.48 24 17.48 24 1.82 1.82 1.82 1.82 1.82 1.82 1.82 1.82	2001 H	3.3 6.3 1.7 2.9001	3,38 ,19 1 ,2,48 ,6,1 1 ,2,48	.846 1.372 .653 54361	92.2 77.1 31.23 95.7 1.73	- 296 - 425 - 294 - 294	1.156 28.86 1.285 1.285	185.3 44.7 194.1	458.7 162.0 162.0 415.7	-536 DAF -967 1	2. 5. 5. 5. 5. 5. 5. 5. 5. 5. 5. 5. 5. 5.	1.885 1.885 2.16.90 2.70.10		016.4	100.5 10000000000	2000 2000 2000 2000 2000 2000 2000 200	25.66 2.66 2.81 5.83	
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MISSIM **VENUS SWINGBY** STOPOVER MISSION WITH INBOUND DATA FOR 2016 OPPOSITION MARS 10. TABLE

316.6-15.0 4.71 62.5 4.7 138.97 161.7 -8.6 29.65 .00000 328.6-14.8 4.74 62.5 4.7 138.97 161.7 -8.6 29.65 .0000 C3-PD C3-PD C3-AD 4.76 138.43 29.65 4.80 138.97 29.65 130.36 27.83 138.97 138.97 29.65 .35 .727 92.5 318.0-14.9 4.75 1.44 1.341 106.8 62.5 4.7 138.97 -3.38 .810 90.4 161.7 -8.6 29.65 - -3.38 .810 50.4 161.7 -8.6 29.65 ETA0 39.73948 1.05169 39.59590 39.47018 39.34483 39.24818 1.05169 39.17862 39.13473 315.8-14.9 62.4 4.4 13 161.7 -8.6 2 160008 315.5-14.7 59.6 4.3 11 164.0-19.5 2 319.3-14.8 62.5 4.7 1 161.7 -8.6 321.8-14.8 62.5 4.7 1 161.7 -8.6 MARS VENUS EARTH PLANET - U PLANET - U PLANET - E 1.04888 ETA4 1.05169 1.05169 1.05169 1.05169 **a** a a ar ar ar .42 .726 92.2 1.44 1.341 106.8 -3.31 .810 99.4 E LEADING EDGE .13 .730 93.3 1.44 1.341 106.8 -3.38 .810 30.4 E LEADING EDGE .21 .729 93.1 1.44 1.341 106.8 -3.38 .810 50.4 E LEADING EDGE .884 --- .29 .728 92.8 .669 --- 1.44 .1341 166.8 --- -1.48 .819 90.4 LIGHT SIDE LEADING EDGE 869 --- 156 .722 92.8 616 --- 1.29 1.342 106.1 --- 3.44 .819 92.1 LIGHT SIDE LEADING EDGE .49 .724 92.3 1.35 1.348 106.8 -3.38 .810 90.4 IDE LEADING EDGE ແທບ DEPARTURE PASSAGE ARRIVAL IS4 IS4 ETA3 2.03876 1.44678 14.21554 7.12764 2.09200 1.44678 14.33243 7.26809 2.10588 1.44673 14.39833 7.27670 2.10588 1.44678 14.54960 7.29516 2.10588 2.10588 2.10588 PERIM APWEL I 2 V 2 PERIM APHEL I 2 V 2 PERIM APHEL I 2 V 2 PASS CONDITION ETA2 6.85547 7.28573 7.38497 LIGHT SIDE LICHT SIDE LIGHT SIDE LIGHT SIDE . 886 . 605 LIGHT SIDE 877 . 869 ----.616 ----ETA1 2.31764 1.42367 14.04837 1.44678 14.47081 14.63445 1 1.166 294.6 533.1 1.021 197.5 422.4 7 227.37 .00 84.92 THET1 THET2 F THET1 THET2 F THET1 THET2 F THET1 THET2 F RAS DECS ETA 1.167 297.2 530.2 1.021 197.5 422.4 61 190.6 179.2 227.37 .00 84.92 -2.0 1.55 1.073 101.7 .247 1.159 293.0 532.9 -29.4 -2.61 .675 100.5 .402 1.019 196.4 422.5 5.457 100.6 158.0 70.7 .334 .761 100.6 179.2 5.457 100.6 168.0 142.48 11.73 227.37 .00 85.00 1.164 296.1 531.2 1.021 197.5 422.4 .761 190.6 179.2 227.37 .00 84.92 -1.5 1.50 1.073 102.0 .251 1.169 292.1 531.6 -298 -2.63 1.676 100.0 .397 1.062 1955 9 420.5 7 -24 1.516 71.6 .325 .769 97.6 175.6 5.443 100.5 166.0 139.77 13.92 223.33 .00 83.75 1.162 295.1 532.1 1.021 197.5 422.4 .761 100.6 179.2 227.37 .00 84.92 1.169 298.2 529.2 1.021 197.5 422.4 .761 100.6 179.2 227.37 .00 84.92 GAM4 1.44678 **GAR3** č.39248 2.44324 2.44324 2.44324 2.44324 2.44324 ECCEN SMA ECCEN SMA ECCEN SMA ECCEN SMA DECP -2.1 1.60 1.074 101.5 .244 -29.6 -2.67 .578 101.2 .404 1.05 1.210 70.7 .334 5.424 100.6 167.4 142.58 12.37 -2.4 1.63 1.075 101.3 .242 -29.5 -2.67 .678 101.2 .404 .1 .05 1.216 5.424 100.6 167.4 142.53 12.37 -2:6 1.67 1.076 101.1 .241 -23.6 -2.67 .678 101.2 .404 .1 .05 1.210 70.7 .334 5.424 100.6 167.4 142.58 12.37 -2.7 1.71 1.077 100.9 240 -29.6 -2.67 101.2 404 1.051 21.210 70.7 334 5.42 100.6 167.4 142.58 12.37

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ORIGINAL PROD

ORIGINAL P. OF POOR QUALITY

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MOISSIM SWINGBY VENUS OUTBOUND **OPPOSITION** WITH 2018 ER MISSION DATA FOR STOPOVER MARS 11. TABLE

TABLE 12. MARS STOPOVER MISSION WITH INBOUND VENUS SWINGBY MISSION DATA FOR 2020 OPPOSITION

	UHP C3-PD C3-AD	ETAD	3.32 67.53 16.50	2488	3.96 67.53 16.60	2101	3.91 67.77 16.54	1140	3.95 37.77 16.54	823	3.99	364	4.03 6.54 6.54	193	6.68 5.68 4.68	883
MARS VENUS FARTH	DECL		18.1 14.5 0000	22.9	0.4 0.4 0.4 0.6 0.6 0.0 0.0 0.0 0.0 0.0 0.0 0.0 0 0.0 0 0.0 0 0.0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	22.9	60.0 60.0 60.0 60.0 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7	22.91	99.0 99.0 99.0 99.0 99.0	22,00	0.4.00 0.1.00 0.00 0.000 0.000	22.91	9.09 1.1 9.09 1.0 1.0 1.0 1.0 0.0 0 1.0 0.0 0 1.0 0 0 0	22.92	90,000 90,000 90,000 90,000	22.93
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and	PSI PSI PSI	TA3	163 81 91 06E	63	163 91 91 91	53	100 100 100 100 100 100 100 100 100 100	5		90	6.863 2.683	9	6888 6888 8	9	0 - 0 U	•
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	I N I N CONDIT	ETA2	-2.25 90 1.77	6688	-2.21 90 1.77 LEA	8581	-2.28 84 1.75 LEA	7329	-2.24 84 1.75 LEAL	7888	-2.20 84 1.75 1.75	890		815	2.23 77 1 1.75 LEAD	268
	APHEL Aphel Aphel Pass	9 1	 ek side	5.4	K SIDE	5.4	 K SIDE	S.S	K SIDE	s.s S	K SIDE	5.58	C SIDE	5.58		S.68
	PERIH PERIH PERIH	ET	.986 612 DAI	9.2829	.988 612 DAF	9.32654	.996 .611 DAR	8066E.6	.992 .611 DAR	.43624	.994 .611 DAR	.48217	-956 	.52907	. 998 .611 DARK	.60351
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SWINGBY MISSION VENUS WITH OUTBOUND DATA FOR 2622 OPPOSITION STOPOVER MISSION MARS 13. TABLE

C3-PD C3-AD DUCP C3-RD .714 --- 7.83 1.262 93.5 144.4 59.6 35.11 --- -2.88 .713 101.6 46.2 7.3 35.85 DAPX SIDE TRAILING EDGE .00000 .845 1.481 .05 1.053 104.2 310.8-18.0 62.46 .714 --- 7.99 1.262 93.6 144.5 60.6 36.25 --- -2.86 .710 101.1 45.5 7.6 34.67 DAPK SIDE TRAILING EDGE .00000 .840 1.483 .09 1.053 104.4 312.3-17.4 64.59 .714 --- 8.16 1.262 93.6 144.6 60.3 37.51 --- -3.84 .708 100.6 44.8 6.7 33.62 DAPK SIDE TRAILING EDGE .00000 .836 1.486 .13 1.053 104.6 313.7-16.5 66.78 .713 --- 7.84 1.262 93.8 145.4 58.5 35.99 --- -2.62 .708 100.7 44.6 7.5 33.58 DARK SIDE TRAILING EDGE .44.6 7.5 33.58 .49000 .13 1.053 104.6 313.7-15.2 66.78 .712 --- 7.99 1.261 93.9 145.5 58.8 37.19 --- -2.44 .710 101.2 45.3 8.5 34.66 DARK SIDE TRAILING EDGE .00008 .840 1.483 .09 1.053 104.4 312.3-17.4 54.59 ETA0 \$60E+.EC 13.48404 33.64709 .712 --- 8.16 1.261 94.0 145.6 59.1 38.42 --- -2.42 .707 100.7 44.5 8.3 33.60 DARK SIDE TRAILING EDGE .00000 .131.053 104.6 313.7-16.9 66.78 93.1 .168 .855 195.5 389.6 .711 --- 7.77 1.261 94.2 146.4 56.9 36.48 85.9 .344 1.093 16.2 157.9 --- -2.23 .707 100.8 44.3 8.9 33.66 14.43 45.71 285.47 .00 89.28 LIGHT SIDE TRAILING EDGE 44.3 8000 33.0 .280 1.161 187.7 439.0 .836 1.486 .13 1.053 104.6 313.7-16.9 F.78 33.30974 33.54429 33.75555 33.47308 DEPAPTURE PLANET - EARTH Passage Planet - Venjs Arrival Planet - Mars R A DEC'. ETA4 3.02903 Z.ZZ144 1.95022 1.69325 1.90317 15.04920 6.67986 Z.55819 1.11031 E7411.1 1.88995 1.78821 1.96925 15.02996 6.86902 2.66161 1.11934 1.91814 1.78886 1.93565 15.82584 6.77112 2.68885 1.11473 αα 1.11934 3.18968 2.24490 1.88940 1.70821 1.96925 15.03656 6.86856 2.66161 1.11934 1.11934 αœ ານຄ 5 1 S d ISd ISd E103 1.88896 1.70821 1.96925 14.98895 6.86820 2.66161 2.66161 THETI THETZ PERIM APHEL I 2 U 2 THETI THETZ PERIM APHEL I 2 U 2 RAS DECS ETA PASS CONDITION ECCEN SMA THEL TI THET2 PERIH APHE I 2 U 2 ETA2 6.86987 ETA1 1.96925 15.02044

 59512.0
 59663.1
 13.36
 313.9
 -60.9
 -5.08
 .917
 91.4
 .165
 .855
 185.1
 385.1

 59663.1
 59816.0
 37.65
 17.41
 24.3
 3.58
 1.361
 37.8
 343
 1.099
 8.5
 155.7

 59663.1
 59816.0
 37.65
 1.43
 3.58
 1.361
 37.8
 343
 1.099
 8.5
 155.7

 5663.1
 5918
 2.516
 50.6
 22.12
 40.13
 277.48
 00
 101.14

 5617.0
 60150
 1.43
 2.856
 110.5
 69.6
 22.12
 40.13
 277.48
 00
 101.14

 5213.6
 60150
 2.2.0
 -1.87
 7704
 92.3
 137.5
 137.55

 5213.7
 1.00222
 1.69322
 3.002303
 2.22144
 1.95022
 1.90325
 1.90317
 11

 92.2 .166 .855 191.1 387.3 86.9 .342 1.094 12.2 157.8 18.54 42.41 281.48 .00 95.21 93.0 .280 1.161 187.7 439.0 92.8 .167 .855 194.2 388.4 86.4 .343 1.093 14.1 157.8 14.96 43.95 283.47 .00 91.07 93.6 .280 1.161 187.7 439.0 92.5 .166 .855 192.7 387.8 86.6 .343 1.096 13.3 156.7 16.22 43.68 282.4 .00 92.71 93.0 .277 1.162 187.8 438.3 GAR4 GAMB 1.70821 ECCEN SMA ECCEN SMA DECP сц Ш 1.89096 PSI 1 PSI 1 RAR P51 1 S e.23867 3.18968 2.2222 2.23256

 59516.0
 59664.4
 14.19
 311.6
 -62.1
 -5.31
 919

 59664.4
 59820.0
 38.05
 173.0
 24.4
 3.61
 1.360

 2.572
 41.0
 .2974
 1.384
 2.856
 110.5
 70.9

 2.572
 41.0
 .2974
 1.384
 2.856
 110.5
 70.9

 5.9880.0
 60154.0
 26.75
 16.5
 -1.1
 -1.89
 609

 7.711
 3.79325
 2.996021
 2.52032
 3.1896.8
 2.23867

 59518.0 59665.6 13.43 309.2 -60.1 -4.92 .919 59665.6 59820.0 38.52 171.6 22.9 3.42 1.359 2.666 6 49.3 .2933 1.406 2.904 110.1 70.8 59880.6 60154.0 26.75 16.5 -111 -1.89 .699 77803 2.99376 2.52032 3.18968 2.22222 59520.0 59666.3 13.90 307.9 -60.5 -5.03 .920 59666.3 59818.0 38.92 170.7 21.5 3.25 1.360 51.1 4. 2995 1.376 2.828 110.7 71.9 59878.6 69152.0 26.47 15.4 -1.6 -1.88 .702 7758 3.06589 2.58001 3.10867 2.23256 **59522.0 59666.9 14.49 305.7 -60.% -5.16 .920 59666.9 59820.0 39.31 170.2 21.5 3.26 1.359 2.451 41.5 .3032 1.366 2.822 110.8 72.5 5980.0 60154.0 26.75 16.5 -11.1 -1.89 .639 7407 2485 2.52032 3.18968 2.3490** 2.22850 I U I I GAM1 Ž ມ 2005 1000 1000 1000 DECL N C3-RS R A CORD2 CORD3 DU2 DU3 ۵ ۵ Ŧ ac ac 00-E0 -E0 -E0 -E0 -DEPARTURE PLANET - EARTH Ph.Sage Planet - VENUS Arrival Planet - Mars PASS ARRIVE XAPPA RESTART RETURN CORD1 DU1 DEP4RT PASS RCP

MARS STOPOVER MISSION WITH OUTBOUND VENUS SWINGBY MISSION DATA FOR 2025 OPPOSITION TABLE 14.

DEPARTURE PLANET - EARTH Passage Planet - VENUS Arrival Planet - Mars ωw PSI PSI THETI THETE PERIM APHEL I 2 U 2 THETI THETE PERIM APHEL I 2 U 2 RAS DECS ETA PASS CONDITION ECCEN SMA ECCEN SMA DECP I U I PSI I I U I PSI I ANMAX INH RAP DECL DECL PASS C3-DD R A ARRIVE C3-PD R A KAPPA VP AH DEPARTURE PLANET + EARTH PASCAGE PLANET + VENUS ARRIVAL PLANET + MARS DEPART PASS RCP

.314 .765 188.2 458.4 .527 --- 3.39 1.265 108.1 125.5 10.4 124.41 387 1282 152 153 5.5 157 5.5 157 21.3 5.7 418.41 8.51 309.79 5.00 55.48 DARK 51D TANLLNG EDG 1.184 1.237 182.0 353.3 1.009 1.455 -.89 1.082 91.4 319.8-26.4 7.95 .315 .770 189.9 458.2 .528 ... 3.38 1.206 108.1 125.7 10.4 124.67 .387 1.024 59.2 168.7 ... -- -.75 .670 97.0 21.3 5.? 40.84 8.47 309.79 .00 95.55 DARK SIDE TRAILING EDGE .00000 .184 1.237 182.0 369.3 1.005 1.465 -.89 1.082 91.4 319.8-26.4 7.55

 25.3
 .315
 .771
 191.7
 453.6
 .528
 -- 3.37
 1.207
 108.1
 125.8
 10.3
 125.15

 74.5
 .387
 1.0624
 59.2
 168.7
 -- -.75
 .676
 97.6
 21.3
 6.7
 40.84

 45.45
 .309.79
 .00
 95.61
 DARK SIDE
 TRAILING EDGE
 .00000

 90.4
 .184
 1.237
 182.6
 369.3
 1.069
 1.465
 -.89
 1.0682
 91.4
 319.8-26.4
 7.95

 60194.0 60357.5 25.16 253.7 -21.3 .27 .824 90.6 .316 .766 181.3 459.3 .524 --- 3.42 1.203 108.2 125.2 10.4 126.20 60357.5 60494.0 122.00 147.6 7.1 2.06 1.331 74.5 .387 1.024 59.2 168.7 --- 7.5 .679 97.0 21.3 0.7 45.84 1.747 22.4 .4554 .421 5.146 101.2 12.1 45.13 8.37 309.79 .00 95.29 DARK SIDE TRAILING EDGE .00060 65544 60810.0 13.11 9.4 -2.3 -.96 .746 90.4 .184 1.237 182.0 369.3 1.009 1.465 -.89 1.082 91.4 319.8-26.4 7.95 1.4440 1.735 1.4349 .80288 2.47434 2.08818 1.36473 1.18599 10.55187 4.23363 1.52181 1.02089 2.6.10893 71.4 .315 .766 183.0 459.1 .525 --- 3.41 1.204 108.1 125.3 14.4 125.32 71.5 .327 1.024 59.2 168.7 --- --75 .670 97.0 21.3 6.7 40.84 45.10 4.46 309.79 .00 95.33 DARK SIDE TRAILING EDGE .00000 90.4 .184 1.237 182.0 369.3 1.009 1.465 -.89 1.082 91.4 319.8-26.4 7.95 92.9 314 768 186.4 458.7 .527 --- 3.39 1.205 108.1 125.5 10.4 124.45 74.5 387 1.024 59.2 168.7 --- --7 .55 .670 97.0 21.3 6.7 40.84 45.28 8.53 349.79 .00 95.43 DARK SIDE TRAILING EDGE .000000 90.4 .184 1.237 182.0 369.3 1.009 1.455 -.89 1.082 91.4 319.8-26.4 7.95 26.32925 C3-PD C3-AD DUCP 26.05683 26.16710 26.56880 26.07701 R A DECL C3-RD 26.88351 ÉTA0 DECL 2.08818 1.36473 1.18599 10.55764 4.23363 1.52181 1.02089 2.08818 1.36473 1.18599 10.54871 4.27363 1.52181 1.02089 2.08818 1.36473 1.18599 10.57371 4.23363 1.52181 1.02089 2.08818 1.36473 1.18599 10.59745 4.23363 1.52181 1.02889 ETA4 2.08818 1.36473 1.18591 10.54663 4.23363 1.52181 1.02089 ≪ ≪ 1.36473 1.18599 10.62863 4.23363 1.52181 1.92089 ECCEN SMA THEL TI THET2 PERIH APHE I 2 U 2 PSI 2 ETA3 ETA2 ETAL CAR4 GANG 2.08818 ĩ 93.7 95.33 95.33 60264.0 60357.5 26.63 244.3 -20.1 .27 .831 94.5 60357.5 60494.0 122.00 147.6 7.1 2.00 1.333 74.5 1.811 21.9 .4519 .424 5.272 100.9 12.3 45.39 60554.0 60810.0 13.11 9.4 -2.3 -.96 .746 90.4 1.4440 3.46570 1.45368 .80288 2.50709 2.088 69 G PSI 1 .80288 2.47055 60196.0 60357.5 24.99 251.8 -21.1 .27 .825 60357.5 60494.0 122.00 147.6 7.1 2.06 1.333 1.764 22.3 .4542 .423 5.172 101.1 12.4 60554.0 60810 13.11 9.4 -2.3 -.96 .746 1.440 1.735 1.4349 .80288 2.47055 4.25717 3.46570 1.46368 .80288 2.47055 .80288 2.47206 60357.5 60434.6 122.80 246.1 -20.3 .27 .829 60357.5 60494.6 122.80 147.6 7.1 2.06 1.333 1.802 22.0 .4520 .424 5.247 101.0 12.4 6554.0 60810.0 13.11 9.4 -2.3 -.95 .746 1.4440 1.7335 1.4349 .80288 2.49007 4.29420 3.46570 1.46368 .80288 2.49007 2.52935 .86288 2.47850 60206.0 60357.5 27.63 242.6 -19.8 .27 .833 60357.5 60434.0 122.00 147.6 7.1 2.06 1.333 1.818 21.8 .4519 .423 .423 100.9 12.3 60554.0 60810.0 13.11 9.4 -2.3 -.96 .746 1.4440 1.733 1.4536 1.46368 2.62935 60200.0 60357.5 25.35 247.9 -20.6 .27 .828 63357.5 60494.0 122.00 147.6 7.1 2.06 1.333 1.791 22.1 .4525 .424 5.222 101.0 12.4 6554.0 60810.0 13.11 9.4 -2.3 -.96 .746 1.440 1.7228 3.46570 1.46368 .86288 2.47850 60198.0 60357.5 25.06 249.9 -20.8 .27 .826 60357.5 60494.6 122.00 147.6 7.1 2.06 1.333 1.778 22.2 .4532 .424 5.196 101.1 12.4 60554.6 60810 1311 9.1 -2.3 -.96 .746 1.446 1.7355 1.4349 .746 4.26003 3.46570 1.46368 2.47206 11 01 GARI ž DECL RM C3-RS RA CORD2 CORD3 DU2 DU3 RESTART RETURN CORD1 DV1

ORIGIDAL OF POCR QUALL

ORIGNUE : OF POOR CO (4)

DEPARTURE PLANET • MARS Passage Planet • Uenls Aariual Planet • Earth	RIM APHEL I 2 U 2 PSI 2 R A DECL 3-PD RIM APHEL I 2 U 2 PSI 2 R A DECL 3-PD RIM APHEL I 2 U 2 PSI 2 R A DECL C3-AD RIM APHES CONDITION	ETAL ETAZ ETAJ ETA4 ETAO	989	79856 6.20181 1.89561 1.04178 31.60467	94075 .703 1A1.5 265-1-26.5 5.189 1.80 1.383 771 8 348.3 7.1 104.95 5523.28 .835 91.0 159.0 -6.2 23.09 554 realLing Edge .00000	85689 6.24838 1.89561 1.04178 31.59108	.990	.92340 6.21525 1.89561 1.94172 31.63176	.99640 .704 101.7 265.5-23 5.94 1.80 1383 77.8 348.3 -7.1 104.95 2.2 28 1835 91.0 150.0 -6.2 23.09 -552328 1835 91.0 150.0 -6.2 23.09 -552	.99397 6.22212 1.89561 1.94178 [.71346	.99626 .764 101.8 265.7-24.8 5.97 1.80 1.383 77.8 348.3 7.1 104.95 -523.28 .835 91.0 150.9 6.2 23.09 .5523.28 .835 91.0 150.9 6.2 23.09 DARK SIDE TRALLING EDGE .000100	.06753 6.22892 1.89561 1.04178 31.83389 	.99013 705 101.8 255.8-24.4 595 11.80 1.383 771 8.343 7.1 104.95 .5523.289 1.383 91.0 150.0 -6.2 23.09 .552 Trailing Edge .0000.00000	.,14338 6.23560 1.89561 1.04178 31.98463		I.22106 6.24212 1.89561 1.04178 32.17045
	EN SIA THETI THETE PE En Sia Theti Thete PE En Sia Theti Thete PE En Sia Theti Thete PE Ecp Ras Decs Eta	GAM3 GAM4	5 1.365 349.5 505.5 . 7 1.197 208.7 319.1 - 5 .782 266.4 537.6 . 7 20.67 .00 106.05	2.07407 1.36462 13.	5 1.366 351.2 505.1 . 7 11.197 208.7 319.1 - 15 .782 266.4 537.6 . 7 20.67 .00 106.05	2.87487 1.36462 13.	6 1.367 352.3 504.8 37 1.197 208.7 319.1 35 .782 266.4 537.6 57 20.67 .00 106.05	2.87487 1.35462 13	76 1.368 354.6 504.6 37 1.197 208.7 319.1 35 .782 266.4 537.6 57 20.67 .00 106.05	2.07407 1.36462 13	77 1.369 356.3 504.3 37 1.197 208.7 319.1 36 7262 266.4 537.6 57 20.67 00 106.05	2.07407 1.36462 14	77 1.370 358.1 504.1 37 1.197 208.7 319.1 95 .782 266.4 537.6 67 20.67 .00 106.05	2.07407 1.36462 14	78 1.376 359.9 563.9 37 1.197 268.7 319.1 95 .782 266.4 537.6 67 20.67 .00 106.05	1 2.07407 1.36467 14
	DECL I U PSI ECCE DECL I U PSI ECCE DECL I U PSI ECCE MARAX INH PSI ECCE	DU4 GAM1 GA M2	36.0 2.69 1.132 92.3 .27 -4.4 -1.58 .662 108.8 .43 2.6 .79 1.212 106.7 .29 .47 106.9 9.2 274.47 7.69	1,46338 2,29843 1,98918	35.1 2.49 1.133 91.9 .27 -4.4 -1.58 .662 108.8 .43 2.6 .79 1.212 106.7 .29 2.47 106.9 1.212 204.47 7.6	1.46330 2.27980 1.91775	34.4 2.33 1.134 91.5 .27 -4.4 -1.58 .662 108.8 .43 2.6 .79 1.212 106.7 .29	1.46330 2.27184 1.92679	33.8 2.15 1.134 91.2 27 -4.4 -1.58 .662 108.8 43 2.6 7.5 1.212 106.7 23 2.47 7.6	1.46330 2.26622 1.93581	33.3 2.06 1.135 20.8 22 -4.4 -1.58 .662 1008 2.6 2.6 .79 1.212 106.7 .23	1.46330 2.26279 1.94471	32.8 1.96 1.136 90.4 .2 -4.4 -1.58 .662 108.8 .4. 2.6 .79 1.212 106.7 .3 3.47 106.9 9.2 274.47 7.3	1.46330 2.26146 1.95344	32.4 1.86 1.136 93.6 .2 -4.4 -1.58 .662 108.8 .4 2.6 .79 1.212 106.7 .2 3.447 106.9 9.2 274.47 7.	1.46330 2.26217 1.96193
DEPARTURE PLANET • MAPS Passace Planet • Venus Arrige Planet • Certh	LALVICH DEPART STOP C3-DD R A DEPART PASS C3-DD R A PASS C3-DD R A DR R A	CORDI CORDZ CORD3	613 49 61554.0 16.66 151.7 612 40 61758.5 40.34 178.4 61758.5 61916.0 106.10 21.9 61758.5 61916.0 106.1 406.9	1.2930 1.7470 1.3769 1.2930 1.7470 1.3769 3.0685 3.04357 3.43379	61346.9 61554.0 16.15 149.9 61614.0 61559.5 40.34 178.4 61614.6 61759.5 40.34 178.4 61759.5 61216.0 166.10 21.9	1.212 JJ. 7.470 1.3759 1.2930 1.7470 1.3759 2.2305 3.66496 3.43379	61348.0 61554.0 15.77 148.2 61614.0 61750.5 40.34 178.4 61614.0 61750.5 40.34 178.4 61750.5 61916.0 106.10 21.9	1,212 33. 1.7455 1.3769 1.2769 1.3769	61350-0 61554.0 15.51 146.4 61614-0 61554.0 15.51 146.4 61750-5 61916.0 106.10 21.9	1.212 33.7 .4558 .456 1.2930 1.7470 1.3769 1.2023 1.0006 3.43770	3.85883 3.19990 3.1996 61352.0 61554.0 15.34 144.6 61614.0 61750.5 40.34 178.4 61750.5 51916.0 106.10 21.9	1.212 33.7 .4550	61354.0 1558.0 15.28 142.8 61614.0 61759.5 40.34 178.4 616159.5 61916.0 106.10 21.9 61759.5 61916.0 106.10 21.9	1.2530 1.7470 1.3769 1.2530 1.7470 1.3769 3.84092 3.15173 3.43379	61356.0 61554.0 15.31 141.0 61614.0 61750.5 40.34 178.4 61614.0 61750.5 40.34 178.4 61750.5 61916.0 106.10 21.9	1.2930 1.7470 1.3769 1.2930 1.7470 1.3769 3.84241 3.17216 3.43379

MARS STOPOVER MISSION WITH INBOUND VENUS SWINGBY MISSION DATA FOR 2027 OPPOSITION TABLE 15.

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MARS STOPOVER MISSION WITH DOUBLE VENUS SWINGBY MISSION DATA FOR OUTBOUND LEG FOR 2029 OPPOSITION 16. TABLE

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C3-PD C3-AD DUCP 54.25 31.15 54.13 30.59 53.95 28.78 53.7**0** 28.38 57.44 289.3-62.8 56.97 217.8 2.1 26.13 .03000 299.1-61.7 56.45 218.0 ..7 27.12 .00000 6.11593 6.00994 5.77392 5.69126 5.66806 5.53443 5.59281 ETA0 288.3-64.3 5 220.1 2.5 2 220.1 2.5 2 287.8-65.2 221.1 1.9 221.1 (9980 288.9-63.3 5 219.6 1.2 2 219.6 0.000 287.3-66.1 221.5 3.4 90000 288.5-63.9 218.7 2.5 218.7 2.5 EARTH UENUS MARS DECL .00000 .00000. PLANET . PLA ETA4 . 00000 .00000. .00000 .00000. .00000 **a** a <u>a</u> a .713 --- -10.46 1.262 93.8 --- -10.46 1.262 95.4 --- -104T SIDE TRAIL.46 EDGE .712 --- -10.32 1.262 94.0 --- 4.59 .609 94.8 --- Light side trailing edge -714 --- -10.05 1.262 93.7 --- --- 4.70 .611 96.6 LIGHT SIDE TRAILING EDGE .711 --- -10.16 1.263 94.2 --- 3.97 .609 96.0 LIGHT SIDE TRAILING EDGE .718 --- -10.34 1.261 93.1 --- 5.73 .614 97.7 DARK SIDE TRMILING EDGE .717 --- -10.26 1.261 93.3 --- -- 5.21 .613 97.8 DARK SIDE TRAILING EDGE .715 --- -10.16 1.262 93.5 --- 5.13 .612 96.6 LIGHT SIDE TRAILING EDGE **ດ** ເບ DEPARTURE Passage Arrival 15d ETA300000 . 98666 . 86668 . 00000 . 00000 .00000. PERIH APHEL I Z U Z PERIH APHEL I Z U Z PASS CONDITION 2.19840 1.08240 2.38454 1.89769 2.30377 1.09138 ETA2 1.09003 2.23801 1.08529 1.89966 1.08573 ETA1 2.41300 2.28338 2.23452 THETI THETZ I THETI THETZ I RAS DECS ETA .857 183.7 382.5 1.201 7.7 167.9 65.98 .00 95.82 .857 185.0 383.8 1.200 9.6 167.9 67.97 .00 92.44 .857 186.3 385.1 1.195 11.7 169.8 69.97 .00 89.65 .857 187.6 386.4 1.194 13.7 169.8 71.96 .00 86.77 .857 189.2 387.0 1.189 14.8 171.6 72.96 .00 84.78 .857 190.5 388.3 1.187 16.8 172.6 74.95 .00 82.40 . 80000 . 68000 .857 191.7 389.6 1.189 18.9 170.9 76.95 .00 80.12 . 68860 Garta . 00000 . 00000 . 00000. . 69669 . 00000 CHAD . 88888 . 00000 . 00000 . 00000 . 00000 . 00000 ECCEN SMA ECCEN SMA DECP 9.69 .917 90.7 .162 -5.56 1.387 87.8 .397 109.7 73.0 163.09-34.95 61852.0 62005.0 27.23 47.4 74.8 8.89 917 91.0 164 62005.0 62006.0 56.34 318.4 -29.1 -5.07 1.387 87.3 398 1.768 40.8 3551 .947 2.867 110.4 73.4 161.04-37.18 .7925 .6853 1.3977 .00000 .00000 2.52038 1.80903 61854.0 62006.3 26.60 48.4 73.9 8.69 .916 91.2 .165 62006.3 62210.0 57.51 316.8 -28.7 -5.06 1.386 86.7 .397 1.810 39.9 .3577 .938 2.929 110.0 73.5 159.53-38.07 .8022 .6709 1.3977 4.32473 2.66394 .00000 .00000 2.59629 1.76113 72.8 8.47 .915 915 .166 -26.2 -4.66 1.386 86.1 .398 2.854 110.5 74.0 157.73-40.20 73.0 8.80 .915 91.8 .167 -26.9 -4.84 1.385 85.8 .396 2.869 110.4 74.6 156.15-39.84 71.8 8.55 .914 92.1 .169 -25.4 -4.65 1.335 85.2 .397 2.867 110.4 74.9 154.88-41.41 70.6 8.30 .914 92.4 .171 -21.7 -4.04 1.386 84.6 .400 2.733 111.5 75.6 153.08-44.27 ĩ 1.75088 .00000 2.51747 1.69203 1.82390 1.71784 PSI 1 PSI 1 RAP ۲ ۵ .00000 2.49247 .00000 2.53458 2.50291 GATI .00000 2003.8 27.86 46.6 75.6 2206.0 57.28 319.8 -31.5 -33.3 .3558 .938 2.973 1 .6853 1.3977 .6853 1.3977 2.82878 .00000 .000 ž DECL DECL EH 61858.8 62088.1 27.77 48.8 7 62008.1 62214.0 59.15 314.8 -2 1.676 40.8 .3889 .897 2. .8166 .6598 1.3977 .8166 .6598 1.3977 4.37399 2.53753 .00000 010.6 26.45 50.7 -212.0 61.33 312.7 -2.9 .3781 888 2 .6649 1.3977 2.54677 .00000 61856.0 62007.5 25.98 49.4 62007.5 62210.0 57.73 315.5 1.741 41.0 .3613 .939 .8118 .6709 1.3977 4.29874 2.63645 .00000 61860.0 62009.4 27.10 49.3 62009.4 62216.0 60.50 313.7 1.662 40.8 .3703 .899 1 .8263 .6554 1.3977 4.34573 2.47552 .00000 ≪ ⊄ ∝ ∞ Ŧ End CORD2 PLANET - EARTH PLANET - UENUS PLANET - MARS 51862.0 62010.6 2 62010.6 62212.0 6 1.540 42.9 .37 .8358 .6242 PASS PASS ARRIVE CORNI DUL 61850.0 62003.8 62003.8 62206.0 1.858 39.3 .685 .7828 .685 .7828 .685 4.37759 2.8287 DEPARTURE PASSAGE APRIVAL DEPAPT Pass RCF

ORICAL ... OF POOR C.

MISSION SWINGBY MARS STOPOVER MISSION WITH DOUBLE VENUS SW DATA FOR INBOUND LEG FOR 2029 OPPOSITION 17.

C3-PD C3-AD DUCP 49.98 33.93 49.96 33.93 **49.97** 33.93 1.46398 3.52606 50.63 33.93 50.12 33.93 ETA0 3.65898 59.24 73.93 3.59391 3.75142 48 84199 33. PLANET - FARS PLANET - LENUS PLANET - EARTH 324.8 21.2 93.2-35.6 .0000 324.8 23.6 93.2-35.6 .00000 224.7 20.0 93:25-3:50 93:00:00 324.7 1⊈.4 93.2-35.6 00000 324.5 18.9 93.2-35.6 .00000 324.4 18.4 93.2-35.6 .00900 DECL 324.3 17.8 93.2-35.6 .00000 ų. ETA4 .00000. . 00000 αα . 09998 . 00000 . 69999 . 00000 αα 87.5 90.5 EDGE លល --- --- 3.45 1.389 87.5 .720 --- -10.54 .916 90.5 DARK SIDE TRAILING EDGE --- --- 3.35 1.389 87.5 .720 --- -10.54 .916 90.5 DAPK SIDE TRAILING EDGE --- --- 3.26 1.390 87.5 .720 --- -:0.54 .916 90.5 DAPK SIDE TRAILING EDGE DEPARTURE PASSAGE PASSAGE PASSAGE PASSAGE PARTUAL --- --- 3.17 1.391 87.5 .720 --- -10.54 .916 90.5 DARK SIDE TRAILING EDGE -- 3.09 1.392 87.5 -- -9.54 .916 90.5 SIDE TRAILING EDGE --- --- 3.01 1.393 87.5 .720 --- -10.54 .916 90.5 Dark Side Trailing Edge ISd ISd ETA3 . 88688 . 86638 .00000. .00000. . 00000 . 00000 --- -- 3.55 1.388 .728 --- 10.54 .516 Dark SIDE TRAILING L Z U Z L Z U Z CONDITION ETAZ 1.05843 1.05843 1.05843 1.05843 1.05843 1.05843 APHEL APHEL PASS (--- ----726 ---DARK SID ETAI 2.00063 2.00678 2.01350 2.02094 2.02911 2.03808 PERIH PERIH A THET1 THET2 P A THET1 THET2 P RAS DECS ETA 1.213 203.6 351.3 .860 338.5 537.3 40.31 .00 91.75 1.216 204.6 351.3 .860 338.5 537.3 40.31 .00 91.71 1.219 205,6 351.3 .860 338.5 537.3 40.31 .00 91.69 GAM4 1.226 207.5 351.4 .860 338.5 537.3 40.31 .00 91.76 1.223 206.5 351.4 .860 338.5 537.3 40.31 .00 91.71 1.230 208.6 351.5 .860 338.5 537.3 40.31 .00 91.83 351.6 537.3 . 00000 . 99998 . 89698 . 30000 . 00000 . 89869 .00000. 1.235 209.6 3 .860 338.5 5 40.31 .00 GANG . 09300 . 00000 . 00000 . 66666 . 88888 . 00000 . 69969 ECCEN SMA ECCEN SMA DECP -3.39 .646 194.4 .403 9.70 1.260 93.0 .162 114.1 77.4 307.92-42.87 -5.8 -3.30 .649 104.9 .404 66.4 9.70 1.260 93.0 .162 2.423 114.4 77.5 307.96-43.19 .652 105.4 .406 1.260 93.0 .162 77.5 307.97-43.50 -3.15 .655 106.0 .437 9.70 1.260 93.0 .162 114.9 77.5 307.94-43.79 -3.9 -3.08 .658 106.5 .409 66.4 5.70 1.260 93.0 .162 2.346 115.2 77.5 307.86-44.07 .661 187.1 .411 1.260 93.0 .162 77.5 387.74-44.32 -2.7 -2.95 .665 107.7 .413 66.4 9.70 1.260 93.0 .162 2.301 115.8 77.5 307.58-44.57 Ē 1.56198 1.75707 1.50190 1.78490 1.50190 1.50190 1.84880 1.50190 1.88586 1.58190 1.50198 1 154 1 154 1 154 5 1.73145 I I C I I I C I ANMAX INH 1.81548 1.92448 GAM1 -5.1 -3.22 66.4 9.70 2.396 114.7 -3.3 -3.01 66.4 9.70 2.323 115.5 . 00000 .000090 . 68888 .00000 .00000. .00000. . 00000 2 -6.4 66.4 2.452 DECL DECL EH -4.5 66.4 2.371 62270.0 62437.0 30.83 163.6 -62437.0 62592.0 47.43 350.6 6 1.471 49.9 3567 1.073 2. 1.1570 2.3966 1.3974 1.1570 2.3966 1.3974 2.80699 1.91445 .000000 62274.0 62437.0 33.44 163.0 -62437.0 62592.0 47.43 350.6 6 1.417 51.0 23906 1.071 2. 1.1570 2.3906 1.3974 2.1574 2.3906 1.3974 2.3974 437.0 32.09 163.3 592.0 47.43 350.6 0.5 -3697 1.072 2 2.3906 1.3974 1.91445 .00000 62264.0 62437.0 27.64 164.6 62437.0 62592.0 47.43 350.6 1.559 48.1 .3613 1.074 1.1570 2.3996 1.3974 1.1570 2.3986 1.3974 2.58392 1.91445 .00000 62266.0 52437.0 28.62 164.3 62437.0 62592.0 47.43 358.6 1.529 48.7 .3634 1.074 1.1570 2.3966 1.3974 1.1570 2.3966 1.3974 2.65308 1.91445 .00000 1.3974 .00000 62268.0 62437.0 29.68 164. 62437.0 62592.0 47.43 350.6 1.500 49.3 .3654 1.074 1.1570 2.3906 1.3974 1.1570 4.23906 1.3974 2.72704 1.91445 .000000 237.0 34.90 162.7 592.0 347.43 350.6 1.5 373741 1.069 2 2.3906 1.3974 2.3906 1.3974 1.91445 .00000 1.3974 .00000 **a** a 1.3974 .00600 1.3974 a a d 60g 4 23-52 23-52 50 23-52 2 PLANET - MARS PLANET - UENUS PLANET - EARTH RCP RARIVE RCP KAPPA CORD1 DV1 62272.0 62437.0 62437.0 5.92.0 1.444 50.5 1.1570 2.39 1.1570 2.39 2.89259 1.914 62276.0 62437.0 62437.0 62592.0 1.390 515 1.1570 2.39 1.1570 2.39 3.08153 1.914 DEPARTURE PASSAGE ARRIVAL

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511.0

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1.05843

2.04792

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TABLE

MISSION STOPOVER MISSION WITH CUTBOUND VENUS SWINGBY DATA FOR 2031 OPPOSITION MARS 18. TABLE

I)(

267.3 -8.3 110.72 203.1 -2.0 27.29 .00010 61.4 28.7 65.94 C3-PD C3-AD DUCP 267.4 -8.5 110.68 203.1 -2.0 27.29 .00000 61.4 28.7 65.94 C3-RD 65.94 268.6 -7.4 117.86 202.3 -2.1 27.61 .538 .987 -2.39 1.206 106.9 268.7 -7.7 115.99 --- 2.16 .590 94.9 201.7 -1.4 26.32 DARK SIDE TRAILING EDGE .000000 .879 1.705 1.81 1.117 102.9 63.1 28.4 67.78 .544 .985 -2.45 1.207 106.5 267.2 -8.9 111.64 --- 219 .592 96.1 203.1 -2.9 27.29 DARK SIDE TRAILING EDGE .000630 .834 1.703 1.93 1.117 102.7 61.4 28.7 65.94 29.64222 29.14087 28.51912 28.37712 ETA0 .547 --- -2.65 1.200 106.3 267.5 -8.E 109.70 --- -2.67 1.502 95.0 202.5 -122 26.01 DARK SIDE TRAILING EDGE .879 1.705 1.81 1.117 102.9 63.1 28.4 67.78 .767 183.5 455.3 .548 --- -2.72 1.209 106.3 267.6 -9.4 109.60 1.168 41.3 173.3 --- 2.27 .592 95.0 202.5 -1.2 26.01 90.28 .00 104.00 DARK SIDE TRAILING EDGE .00000 1.292 207.9 417.4 .879 1.705 1.81 1.117 102.9 63.1 28.4 67.78 28.31611 28.23781 28.33579 .536 .988 -2.32 1.296 107.0 268.6 -7.4 1 --- 2.09 .590 96.0 202.3 -2.1 DARK SIDE TRAILING EDGE .00000 .884 1.703 1.93 1.117 102.7 61.4 28.7 DEPARTURE PLANET - EAFTH Passage planet - UENUS Arrival Planet - Mars רר מכנר R A DECL ETA4 1.37560 1.95636 11.66110 5.39553 2.54136 1.11757 ¢ ⊄ ≪ ∢ 1.39148 1.98480 11.58102 5.68621 2.58735 1.12149 1.11757 1.11757 1.37560 1.95636 11.47946 5.59001 2.54136 1.11757 1.39148 1.98480 11.44191 5.68075 2.58735 1.12149 1.12149 .546 .985 -2.59 1.208 106.3 --- 2.19 .592 96.1 DARK 5IDE TRAILING EDGE .884 1.703 1.93 1.117 102.7 សស PSI 2 .545 .985 -2.52 1.207 106.4 --- 2.19 .592 96.1 DARK SIDE TRAILING EDGE .884 1.703 1.93 1.117 102.7 PSI PSI ETA3 2.54136 1.37560 1.95636 11.48550 5.59001 2.54136 2.58735 THETI THETE PERIM APHEL I Z U Z Theti Thete Perim Aphel I Z U Z Ras decs eta Pass comdition ECCEN SMA THEL TI THET2 PERIH APHE I 2 U 2 ETA2 1.37560 1.95636 11.49958 5.59601 5.68075 ETAI 1.39148 1.98480 11.45162 762 171.5 457.5 1.166 44.3 172.1 92.78 .00 102.13 1.293 207.1 416.4 763 173.2 457.2 1.164 43.6 173.5 92.39 .00 102.75 1.292 207.9 417.4 764 176.3 456.2 1.170 42.0 171.8 90.77 .00 103.40 1.293 207.1 416.4 .765 178.1 456.0 1.170 42.0 171.8 99.77 .00 103.39 1.293 207.1 416.4 .765 179.9 455.7 1.170 42.0 171.8 90.77 .00 103.38 1.293 207.1 416.4 .766 181.7 455.5 1.168 41.3 173.3 90.38 00 104.01 1.292 207.9 417.4 GAN4 GANG ECCEN SMA ECCEN SMA DECP MA 297 2.433 2.68 2.68 2.12 . 289 . 429 1 . 89

 62526.0
 62693.1
 24.99
 29.6
 22.7
 1.96
 851
 89.2
 287

 62693.1
 62880.0
 110.51
 300.6
 -9.1
 -2.37
 1.382
 77.7
 429

 61197
 32.9
 4725
 -473
 3.532
 106.4
 9.2
 194.17
 429

 62940.0
 1.456
 1.3.56
 160.3
 8.9
 -.09
 .675
 101.4
 .317

 1.4681
 1.4364
 3.651
 8.9
 -.09
 .675
 101.4
 .317

 1.4681
 1.4364
 3.461
 8.45
 8.45
 .367
 .317

 1.4681
 2.55887
 1.56959
 3.15877
 2.47069
 1.72226

 22:1
 1.95
 .851
 89:9
 286

 -9:1
 -2:37
 1.382
 77.7
 .429

 3:545
 196:4
 9.2
 194:16
 .96

 3:545
 196:4
 9.2
 194:16
 .96

 8:0
 -.09
 .679
 101.4
 .317
 21.4 1.95 .852 90.7 .286 91.6 -2.4 1.382 78.0 .427 536 -2.4 1.382 78.0 .427 8.2 .06.4 .680 101.8 .326 8.2 .04 .680 101.8 .326 62532.9 62691.1 25.16 24.1 20.6 1.94 852 91.4 285 62693.1 62884 0 107.76 300.8 9.6 2.47 1.382 78.0 467 62693.1 62884 0 107.76 300.8 9.6 6.4 1.382 78.0 46 62942.6 532.7 4677 481 3.548 106.4 9.8 194.39 1.04 62942.6 5384 0 14.20 161.7 8.2 0.04 680 101.8 320 1.4681 1.3956 31.5504 3.22670 2.47439 1.68893 3.22670 2.46793 1.68893 2 E 1.73052 1.69699 3.15877 2.48002 1.72226 3.15877 2.46668 1.72226 2 1.96 .849 87.2 . -2.46 1.381 77.3 . 105.9 9.6 195.15 2 . .64 .688 161.8 . 23.3 1.97 .851 88.5 -9.1 -2.37 1.385 77.7 3.519 106.5 9.1 194.18 8.0 -.09 .679 101.4 1.92 .849 86.4 -2.30 1.381 77.0 105.9 8.5 194.91 -.09 .679 101.4 PSI 1 PSI 1 RAP PSI 1 5 3.15877 2.54197 3.22670 2.51626 I I U I I I U I ANNAX INH 11 01 Geni 2 62520.0 62694.4 28.19 38.0 24.6 67694.4 62880.0 118.02 300.7 -8.5 1.17 31.8 4829 443 3.655 6290.0 63182.0 13.56 160.3 8.0 1.4744 1.4364 1.50099 3.15 4.39131 2.58140 1.50099 3.15 DECL DECL EH DECL 62524.0 62693.1 25.42 31.5 2 62693.1 62880.0 110.51 300.6 -1.186 33.0 .4740 .471 3. 62940.0 63182.0 13.56 160.3 1.4681 1.4364 1.5887 1.50099 CORD3 CORD3 DU3 6252.0 62694.4 27.04 36.3 62694.4 62884.0 115.01 300.9 1.199 31.8 .4781 .45 62944.0 63184.0 14.20 161.7 1.4744 1.3956 1.55504 62528.0 62693.1 24.81 27.7 62692.1 62880.0 119.51 300.6 1.207 32.8 .4713 .474 62940.0 63182.0 13.56 160.3 1.4681 1.4681 1.468 4.24978 2.55887 1.50099 62530.6 62693.1 24.87 25.9 62693.1 62884.0 107.76 300.P 1.220 32.9 .4683 .481 62944.0 63184.0 14.20 161.7 1.4681 1.3956 .8111 4.25216 2.46688 1.55504 e⊂e⊂ Carc Η CORD2 CORD2 DU2 PLANET - EARTH PLANET - UENUS PLANET - MARS 29-F2 C3-F2 C PASS ARRIUE Kappa RESTART RETURN CORD1 C DU1 DU DEPARTURE | Passage | ARRIUAL | DEPART PASS RCP

ORIGINAL DE ORIGIN

TABLE 19. PLANETARY MISSION STAGE SIZING FOR VENUS SWINGBY OPPOSITIONS 2001 TO 2010

A

PDAG TI S CUTING THPO MD

	11 0.1111		11 0.1111		11.9.1111		1111.0 1		1111 0.1111
	2871. 1470725. 480.000 0.1111		7020. 2858094. 480.000 0.1111		2424. 1953675. 480.000 0		3171. 4664865. 488.068 6.1111		3115. 2487593. 4887593. 0.1111
	L 203 L 203		E O O O O O O O O O O O O O O O O O O O		E C 8 M E C 8		612P3		E 03 1283 63 1383
	25844. 86894. 480.000 0.1111		63183. 175893. 480.000 0.1111		21820. 126205. 480.000 0.1111		28546. 271962. 488.888 8.1111		28836. 166219. 489.998 4.1111
	1592 1592 1592		uF3 UB01 G2 P2 G2 P2		61801 61801 6282 6282 6282 6382 6382 6382 6382 6382		62 1880 1882 1882 1882		UF3 1591 62 22
u I	28715. 782127. 488.000 1.23850	ILE	74203. 1583196. 480.000 2.50300	ILE	24244. 1135962. 480.060 1.07080	ų.	31718. 24473. 480.06 1.34690	64	31151. 1496122. 480.000 1.32670
Y PROFIL	USTG3 UF1 ISP1 Du4	V PROF]	USTG3 UF1 15P1 DU4	84 PR0F1	ust63 uF1 ISP1 DV4	Y PROFIL	USTG3 UF1 ISP1 DV4	PROFILE	UF: UF: DUA
ND SUINCE	420001. 869021. 1.00 3.23879	DUND SUING	42000. 1759889. 1.80 2.27400	IDHINS CHING	42000. 1262168. 1.49640	ND SUINCE	42000. 2719261. 1.00 4.20400	E SUINGRY	42000. 1662341. 1.00 3.82550
INBO	uPROB2 uSTG1 C4 DUS2	00110	uppo b e ustg1 c4 dvs2	OUTBO	4 PROB2 45761 64 DUS2	INBOL	UPROB2 USTG1 C4 DUS2	DOUBI	UPROB2 USTG1 C4 DVS2
2001	9. 35296. 1.98	2003	6. 86873. 1.06 3.55488	2005	44722. 44722. 1.00 3.26530	2067	.09379. 1.00379. 3.18240	2010	6. 57405. 1.00 2.14050
DATE	UPROB 1 U BO2 C3 DUS1	DATE	uPR011 UB02 C3 DUS1	DATE	UPROB1 UB02 C3 DUS1	DATE	LPROB1 UF 72 C _ DUS1	DATE	UPROB1 UB02 C3 DUS1
	95009. 317693. 1.00		95000. 727930. 1.00 0.00000		95068. 402541. 1.00 3.00000		95888. 384588. 1.68 8.68606		95080. 516696. 1.00
			1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		DU DU DU DU DU DU DU DU DU DU		1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1
	83000. 352989. 1.00 3.57210		83000. 88883. 1.69 3.89618		83000 447263 447263 449980		83000. 1093887. 1.00 4.33770		83000. 574101. 1.00 4.33010
	UPL USTG2 C1 DV1		uPL USTG2 C1 DV1		uPL USTG2 C1 DV1		upt USTG2 C1 DV1		upt ustg2 001 D01

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TABLE 20. PLANETARY MISSIONS STAGE SIZING FOR VENUS SWINGBY OPPOSITION 2012 TO 2020

				DATE	: 2012	OUTB	NINS GNNO	GBY PROF	311.						
us 162 011 011	83000. 495227. 1.00 4.38600	LE L	95060. 445788. 6.0000	UPROB1 UB02 C3 DVS1	49518. 1.00 2.65130	UPROB2 USTG1 C4 DUS2	42000. 1647219. 1.20 1.81879	USTG3 UF1 ISP1 DV4	83384. 1482512. 480.000 2.82310	62 12 12 12 12 12 12 12 12 12 12 12 12 12	75046. 164707 480.000 0.1111	103 100 122 132 130 130 130 130 130 130 130 130 130 130	8338. 2445829. 480.000 0.11:1	61.0.11	11
				DATE	2014	OENI	UND SUING	BY PROFI	LE						
uste2 U1 DU1	83 600. 365389. 1.88 3.69598	500 500 500 500 500 500 500 500 500 500	95888. 328781. 1.88 8.888	UPROB1 UBO 2 C3 DUS1 DUS1	3 5528. 1.8638 1.48538	UPROBE USTG1 C4 DUSE	42000 941438 1.000 3.44020	USTG3 UF1 ISP1 DU4	30939. 847303. 488.003. 1.31910	uf3 1801 62	27845. 94135. 480.000	603 15 15 15 15 15 15 15 15 15 15 15 15 15	3094. 1557686. 480.003 0.111	61 0.11	11
				DATE	2016	NB NI	ADN SUING	IV PROFI	u.						
UPL USTG2 C1 DV1	83 000 . 774196. 1.00 4.62990	UNEN UF2 CC2 DV0	95000. 696783. 1.00 8.0000	UPROB1 UBO2 C3 DVS1	77413. 1.00 2.31250	uPROB2 USTG1 C4 DVS2	42000. 2370706. 1.09	USTG3 UF1 ISP1 DU4	43352. 2133657. 480.000 1.73850	0116 0116 011 011 011 011 01 01 01 01 01 01 01 01	29817 237849 488.686 488.686	600 60 60 60 60 60 60 60 60 60 60 60 60	3408254 3408254 486,306 6-1111	6: 0.1:	::
ļ				DATE	2018	0UTBC	DNINS QNIN	BV PROFI	LE						
ur ustez c1 DV1	83000. 403517. 1.00 4.29560	LT CC CC CC CC CC CC CC CC CC CC CC CC CC	95800. 363169. 1.02 0.88000	LPROB1 LBOZ C3 DUS1	48348. 48348. 1.88 2.88938	UPROBE USTG1 C4 DUSE	42000. 1328670. 1.00 1.47680	USTG3 UF1 ISP1 DV4	45827. 1195815. 480.000 1.81650	62 15 15 15 15 15 15 15 15 15 15 15 15 15	41245. 132855. 480.000 0.1111	E C8 1 1 S C8 1	4582. 1998915. 480.000	61 8.111	11
ġ				DATE	2020	INBOU	ND SUINCE	/ PROFIL	ίω						
USTG2 C1 DV1	83000. 383401. 1.00 4.17480	UNE CERC DVG DVG	95999. 345865. 1.00 8.00000	uprob1 ubo2 c3 dvs1	6. 38337. 1.6040 1.67040	498082 45161 64 7052	42000. 1188768. 1.00 3.39040	usTG3	27191. 1069902. 480.000 1.18220	6691 6792 6792 6792	24472. 118866. 480.000		27:9. 18:9361: 480.000	61 0.:11	1

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PLANETARY MISSION STAGE SIZING FOR VENUS SWINGBY OPPOSITION 2022 TO 2031 TABLE 21.

G1 0.1111 31 0.1111 G1 0.1111 GI 0.1111 G1 0.1111 19415. 2833479. 489.849 1751. 2084630. 480.020 9.1111 3504, 2783209. 480.000 4901. 2513434. 480.890 10196. 2275570. 480.000 0.1111 E edst 15P3 15P3 63 90148. 173132. 480.000 0.1111 15762. 137853. 488.088 31539. 173577. 480.000 44115. 167189. 480.000 91771. 158371. 486.000 10801 10801 10801 LF3 LB01 62 22 UF3 UB01 G2F2 G2F2 ura 11572 6272 199164. 1558341. 489.899 3.18970 17513. 1240799. 480.800 0.80290 49917. 1564851. 488.698 1.91458 35843. 1562356. 480.800 1.46330 181967. 1353478. 488.899 OUTBOUND SUINGBY PROFILE OUTBOUND SUINGBY PROFILE INBOUND SUINGBY PROFILE OUTBOUND SUINGBY PROFILE USTG3 UF1 ISP1 DU4 DOUBLE SUINGBY PROFILE USTG3 UF1 ISP1 DU4 ustga ufi Ispi Du4 42000. USTG3 1503240. UF1 1.00 ISP1 1.55500 DU4 USTG3 UFI ISP1 DU4 WPROB2 42000. USTG1 1731473. C4 1.50 DUS2 2.52030 42000. 1378652. 1.00 1.46370 42800. 1735927. 1.00 3.43380 42000. 1672040. 1.60 2.80700 UPROB2 USTG1 C4 DUS2 uPROB2 USTG1 C4 DV52 UPROB2 USTG1 C4 DUS2 UPROB2 USTQ1 C4 DVS2 6. 46842. 1.69 3.46578 79217. 1.60 3.06500 57233. 1.00 2.63650 44972. 1.00 2.46698 DATE 2022 DATE 2025 DATE 2027 DATE 2029 DATE 2031 UPROB1 UBO2 C3 DUS1 UPROB1 UBO2 C3 DV51 uprosi Uboz C3 DVS1 WFROB1 WBO2 C3 DVS1 uPROB1 UBO2 C3 DVS1 0. **1** 95000. 421623. 1.00 0.00000 259665. 783665. 95000. 713022. 1.00 515144. 1.99 95990. 4847990. 1.98 No. No. UFR DVO 83000. 781842. 1.00 3.75850 83000. 468466. 1.00 83000. 792239. 1.00 3.87900 83000. 572377. 1.00 4.29820 1.2529 45762 01 01 NC COL 801 851 62 201 851 62 201 85 20155 20155 UST02 C1 DVI

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TABLE 22. MARS CONJUNCTION CLASS STOPOVER MISSION, MISSION DEFINITION

						_	
TOTAL MISSION TIME (DAYS)	866	9 9	1004	986	28	8	906
INBOUND TRIP TIME (DAYS)	216	700	270	280	1 Se	332	350
MARS STOPOVER TIME (DAYS)	200	550	230	340	340	340	946
OUTBOUND TRIP TIME (DAYS)	282	200	204	356	340	318	906
ARRIVE EARTH	3858 (.1202)	4656 (.1013)	5512 (.1128)	6276 (.0957)	7036 (.0968)	7802 (.1127)	8564 (.1318)
LEAVE MARS	3642 (.0826)	4456 (.0996)	5242 (.1202)	5986 (.1046)	6732 (.0922)	7470 (.0870)	8214 (.0866)
ARRIVE MARS	3142 (.1166)	3906 (1111)	4712 (.0876)	5646 (.0947)	6392 (.0633)	7130 (.0836)	7874 (10940)
LEAVE EARTH J.D. 246000 (EMOS)	2860 (.1107)	3706 (.1016)	4508 (.1091)	5290 (.1363)	6052 (.1193)	6812 (.1060)	7568 (.1003)
EARTH- MARS OPPOSITION J.D. 2460000	2989	3775	4584	5382	6157	6920	7688
MISSION FOR OPPOSITION YEAR	2031	2033	2035	2037	2040	2042	2044

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MARS CONJUNCTION CLASS STOPOVER MISSION,

TABLE 23.

MISSION DATA FOR 2031 OPPOSITION

ARTUR PLANET - EARTH

DEFARTURE PLANET . EARTH AGRIVAL PLANET . HARS

0.568 1.406 -1.72 8.767 87.1 312.4-34.1 3.45 --- -1.61 1.111 88.1 93.8 8.4 12.93 0.970 1.405 -1.84 0.768 87.0 312.4-30.9 3.46 •.971 1.405 -1.95 •.769 86.9 312.4-31.7 3.47 --- -1.42 1.111 88.1 94.2 10.1 12.82 93.3 0.182 1.188 338.0 553.8 0.973 1.404 -2.13 0.769 87.0 313.5-32.7 3.47 96.6 0.242 1.323 201.9 350.2 --- -1.42 1.111 88.1 94.2 10.1 12.82 0.274 1.404 -2.26 0.770 85.9 313.4-33.5 3.49 --- -1.41 1.111 88.4 92.9 10.2 12.64 30 6.02825 3.92661 1.56039 1.02747 13.16961 6.02874 3 2654 1.55956 1.02735 13.13569 6.03401 3.92761 1.55897 1.02732 13.11559 •.275 1.403 -2.39 •.771 86.8 313.4-34.4 3.51 --- -1.32 1.112 88.4 93.1 11.0 12.60 0.877 1.403 -2.60 0.772 86.9 314.5-36.6 3.53 6.05825 3.93006 1.55592 1.02709 13.10819 6.03276 3.92824 1.55897 1.02732 13.19601 6.06646 3.93276 1.55652 1.02703 13.12432 6.08261 3.93414 1.55652 1.62703 13.16087 E TAO U Z PSI Z R A DECL U Z PSI Z R A DECL ETA ELTA PERIH APHEL 1 2 PERIH APHEL 1 2 ETAZ ETAL SNA THETT THETZ SNA THETT THETZ 1.187 333.6 552.9 1.322 200.2 350.4 1.187 334.9 553.5 1.322 201.0 350.3 1.188 336.3 554.1 1.323 201.9 350.2 1.01918 2.17361 1.33797 1.19412 1.24176 93.1 0.181 1.185 339.4 554.4 96.8 0.242 1.323 203.6 351.9 1.02141 2.17883 1.33629 1.19347 1.24235 1.01918 2.17032 1.33937 1.19412 1.24176 1.189 340.8 555.6 1.324 203.5 351.8 1.19319 1.24368 1.01159 2.16655 1.34193 1.19616 1.23976 1.190 342.5 554.7 1.324 293.5 351.8 1.00071 2.16369 1.34820 1.19719 1.23926 1.00071 2.16343 1.34513 1.19719 1.23926 ž ŝ U I PSI I ECCEN 2.89 1.691 53.8 6.183 2.98 0.698 96.4 0.242 2.96 1.692 93.6 0.183 2.96 0.699 96.6 0.242 2.82 1.001 94.0 0.184 3.07 0.697 9.12 0.242 92.9 0.180 97.1 0.243 3.43 1.092 92.6 0.179 2.84 0.708 97.1 0.243 1.62419 2.18465 1.33517 Ì 3.10 1.022 8.90 0.592 3.17 1.092 2.92 0.701 3.25 1.092 2.84 0.702 Ē --ž 12.5 19.5 1.91 1.91 20.02 • - - 61 16.4 17.5 18.8 19.7 18.8 62856.0 63140.0 11.30 227.9 63640.0 53858.0 6.03 200.1 1.2461 1.4598 3.66572 1.36453 0.83249 62054.0 63134.0 11.54 227.1 63638.0 63859.0 6.02 201.1 1.2349 1.4694 3.67828 1.36859 0.83146 62860.0 63142.0 10.88 229.1 63642.0 63858.0 6.06 199.2 1.2573 1.4295 0.83585 3.64729 1.37537 0.83565 62864.0 63146.0 10.54 230.1 63846.0 63869.0 6.19 197.9 1.2706 1.3805 0.84716 3.64233 1.38558 0.84716 5256.0 53145.0 10.55 239.5 5366.0 5366.0 10.55 239.5 1.2785 1.3955.1 1.2785 1.3955.1 2.5259 1.46536 0.84715 62862.0 63144.0 10.70 229.6 33644.0 63860.0 6.14 198.9 1.2685 1.4208 3.63913 1.38436 0.84287 62858.0 63142.0 11.04 228.5 63642.6 63858.0 6.06 199.2 1.2573 1.4295 3.65443 1.37046 0.83505 ≪ « ≌ œ S ARRIVE C3 RETURN C3 D1 CORD2 D4 Condition IN TANET

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TABLE 24. MARS CONJUNCTION CLASS STOPOVER MISSION, MISSION DATA FOR 2033 OPPOSITION

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DEPARTURE PLANET - EARTH MORTUAL PLANET - HARS

DEPARTUNE PLANET . EARTH ARRIVAL PLANET . INARS

• 64556.0 8.88 775 91.7 51.15 1.192 25.1 4.19 0.775 91.7 353.7 4.8 3.31 • 64556.0 8.88 775 91.7 9.102 1.192 25.1 341.7 9.103 1.227 1.89.7 348.6 1.0 0.775 91.7 353.7 4.8 3.31 1.20 343.6 1.20 343.6 1.20 343.6 1.20 343.6 1.20 343.6 1.20 343.6 1.20 343.6 1.20 343.6 1.20 343.6 1.20 <

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TABLE 25. MARS CONJUNCTION CLASS STOPOVER MISSION, MISSION DATA FOR 2035 OPPOSITION

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ARTURE FLANET + EARTH MRIUAL PLANET + NARS

DEPARTURE PLANET + EARTH ARRIVAL PLANET + MARS

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TABLE 26. MARS CONJUNCTION CLASS STOPOVER MISSION, MISSION DATA FOR 2037 OPPOSITION

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DEPARTURE PLANET + EARTH ARRUAL PLANET + MARS

DEPARTURE PLANET - EARTH Araival Planet - Mars

DEPART DE PART	ARRIVE SETURE	55	₫. Œ.	DECL	44 44	-11 >>	PSI 1 PSI 1	ECCEN	SMA	THE T 1 THE T 1	THET2 THET2	PERIH A PERIH A	PHEL	លល មម	ณณ วว	51 2 2 15	αα αα	DECL	d HD C C C C C C C C C C C C C C C C C C C
CORDI	L CORDE	a l		Ž		Cons	GAH	Q	GANG	g	4 5	ETAL	ET	A2	ETA3	w	TA4	ETA	0
5284.0 6	5638. ¢ 1	7.15	101.5	-20.7	-2.51 @.73	1.110	93.8 85.8	9.254	1.350	345.0	561.1 372.5	1.006 1 1.003 1	683. 414.	3.14 9-2-41 1	.684	83.1 91.8	71.5 193.8-	24.0 28.3	2.76 8.10
2.462	3 1,5443 9,97988		66019	.8096	3	2.3009	1 1.2	E+1E	1.29322	1.1	8768	S.71366	3.9	7589	1.511	1.	. 82168	13.1	4660
65286.0	55642.0 11 56274.0 5	6.78 9.98	331.2	- 2 0 -2-	-2.46	1111	92.8 86.1	0.255 0.178	1.351 1.209	346.2 160.3	562.2 373.6	1.007 1 1.003 1	.414	3.15	. 685 . 076	85.8 92.8	72.4	2.53. 4. (5)	2.79 8.10
3.90638	5 1.5253 9.99393	-	18538	.809	57	1665.5	0.1 0.1	113511	1.28647	1.1	18767	5.71007	6.E	6559	1.510	30 1	.02108	13.0	E6E6
65288.0	65644.0 1	6.42 9.79	4.000 4.800	-21.0	-2.45 9.65	91.112	92.5 86.3	0.254	1.352	347.7	562.6 373.3	1 600.1 1 600.1	. 695 . 414	3.22	0.686 1.076	82.7 91.9	73.6 193.5-	24.4 27.8	2.80 8.11
	7 1.60235		17636	.883	84	2.2866	S 1.6	3732	1.28395	1.1	18774	5.71961	5.E	16208	1.509	* 5	. 82189	13.6	5819
65292 . •	65646.0 1	6.23 9.78	329.6 3 8 7.9	-51.4	-4 -4	3 1.116 9.775	88.4	0.254	1.352	349.2 162.0	563.1 374.7	1.063	626	62.5-	8.687 1.076	82.6 92.1	73.5	8.4.8 9.4.8	2.82 8.13
2.491	7 1.5060 9 1.01196	-	16849	.810	61	3.2814	17 1.6	23985	1.28178	8	18793	5.71505	3.E	15971	1.589	81 1	. 82111	13.0	1785
65292.0	65646.0 1 66276.0	6.13 9.7 9	328.2 9.7	-22.7 3.5	-2- -2:0	5 1.196 2 0.775	91.8	0.254	1,352	351.9 162.0	563.0 374.7	1.0091	414	3.45	0.687 1.076	82.6 92.1	192.7	25.0 -25.4	2.83 8.13
2.491	7 1.5660 9 1.01832	-	16849	.810	161	2.2796	1.1	24153	1.28171	1	18793	5.7222	3.E 7	96946	1.505	81 1	.82111	13.8	4269
65294.0	65646.0 1 66276.0	16.15 9.70	326.7 387.9	-24.1 3.5	2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	2 1.11	91.5 86.4	0.170	1.352	352.8 162.8	562.9 374.7	1.619	1.695	-2.27	0.687 1.076	82.7 92.1	74.2	-26.4	2.85 8.13
3.8788	1.02826	-	.16849	. 816	161	2.279	76 1.	24415	1.2817	8	E8781	5.7364	е В	96163	1.585	81 1	.02111	13.6	175B
65296.0 65236.0	65646.0 66276.0	16.31 9.70	325.1 307.9	-25.6 3.5	00 00	2 0.77	1.16 5	0.253 0.178	1.209	354.6 162.0	562.7	1.010	1.694	3.83 -2.27	0.687 1.076	92.1	74.5	-28.5 -26.4	2.88 8.13 8.13
2.491 3.8855	17 1.5064 1.04231		.16849	0.816) 61	2.283	17 1.	24787	1.2817	8 1.	18793	5.7584		96639	1.505	1 180		1.61	14769

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TARLE 27. MARS CONJUNCTION CLASS STOPOVER MISSION MISSION DATA FOR 2040 OPPOSITION

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BEPARTURE PLANET - EALTH MARIUNE PLANET - EALTH

8.45 8.37 8.45 8.40 8.37 5.20713 3.85085 1.50414 1.02142 11.81193 58 58 5.30403 3.85097 1.50414 1 V2142 11.76833 5.30426 3.85129 1.50414 1.02142 11.74241 5.30875 3.85144 1.50386 1.02141 11.72911 8.31 91 91 5.31611 3.85241 1.50355 1.02136 11.72867 5.32661 3.85440 1.60356 1.02135 11.74177 ETAO 6.34046 3.85744 1.50372 1.02135 11.76894 DEPARTURE PLANET - EARTH ARRIVAL PLANET - MARS 1.0011.646 2.73 0.658 83.3 197.1 18.5 1.014 1.484 --0.62 1.083 90.3 343.9-14.4 1.001 1.645 2.81 0.698 83.4 197.9 18.9 1.014 1.484 -0.62 1.083 90.3 343.9-14.4 1.002 1.644 2.90 0.698 83.4 146.6 19.4 1.014 1.484 -0.62 1.083 90.3 343.9-14.4 1.003 1.644 2.92 0.700 83.3 198.6 19.4 1.014 1.484 -0.56 1.083 90.2 343.9-13.6 1 DECL DECL 1.003 1.644 2.93 0.701 83.1 198.6 19.4 1.014 1.485 -0.46 1.083 90.5 242.3-13.1 1.003 1.644 2.94 0.782 82.9 198.6 19.4 1.014 1.485 -0.40 1.083 90.5 342.3-12.3 1.014 1.486 2.95 0.703 22.3 198.5 19.4 1.014 1.486 -0.32 1.083 00.5 342.3-11.4 € € ¢ ¢ ETA4 C 2 PSI 2 V 2 PSI 2 ETA3 ~~~ E1A2 PERIH APHEL PERIH APHEL ETAL SM THETT THETE 1.323 345.6 561.8 1.249 155.4 361.8 1.323 347.4 561.7 1.248 155.4 361.8 10.0 -1.44 1.11 92.1 0.243 1.323 349.2 561.5 6.4 -1.27 0.758 84.6 0.108 1.249 155.4 361.8 1.323 350.8 562.1 1.249 156.3 361.6 1.1906S 1.19266 1.22952 1.19965 1.324 352.5 562.8 1.249 157.2 363.2 1.324 354.1 563.4 1.850 158.2 363.4 1.22952 1.19865 1.22930 1.19854 1.324 355.7 564.0 1.250 159.1 362.0 1.22976 1.19013 CAN4 0.81805 2.20436 1.19888 1.23046 1.19804 0.81877 2.20373 1.20114 1.23168 1.18999 0.82138 8.23567 1.19236 1.22952 92.5 0.243 84.6 0.188 U I PSI I ECCEN U I PSI I ECCEN 92.3 0.244 84.6 0.128 16.6 -1.43 1.11 91.8 0.842 5.7 -1.33 0.75: 84.8 0.188 10.1 -1.42 1.112 91.5 0.242 4.8 -1.42 0.756 84.9 0.189 91.2 0.248 85.1 0.189 10.3 -1.39 1.113 82.8 0.242 3.3 -1.55 0.753 85.3 0.189 0.82138 2.21377 1.12338 0.81929 2.ñ6625 1.19684 0.82092 2.20.39 1.19501 Š 2.21885 11.5 -1.31 1.110 6.4 -1.27 0.758 10.8 -1.37 1.110 6.4 -1.27 0.758 10.2 -1.41 (.112 4.1 -1.48 0.754 Ī ----0.82138 ł DECL 66044.0 66382.0 13.56 97.2 66728.0 67034.0 7.53 44.5 2.3221 1.8734 9.97256 3.76584 0.82813 0.97256 66946.0 66388.0 13.24 95.9 66728.0 67034.0 7.53 44.5 2.3221 1.8734 7.53 44.5 3.75139 0.82930 0.97256 ≪ « ≪ œ 66448.0 66388.0 12.99 94.5 66728.0 67034.0 7.53 44.5 2.3221 1.8734 3.74060 0.87346 0.97256 66650.0 65390.0 12.78 93.4 66730.0 67034.5 12.78 93.4 2.3300 1.8648 3.73128 0.83857 0.97173 66962.0 66392.0 12.63 92.3 66732.0 67036.0 7.54 43.4 2.3376 1.8560 7.54 43.4 3.72459 0.84577 0.97348 66654.0 66394.0 12.54 91.2 66734.0 67036.0 12.57 42.7 2.3450 1.8471 7.57 42.7 3.72055 0.35377 0.97617 64056.0 66396.0 12.51 90.0 66736.0 6736.0 12.51 90.0 2.3522 1.8399 3.71922 0.86365 0.95482 Eng 9.95682 C0402 C0405 C0405 NAUT 34 RESTART RE CORDI

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TABLE 28. MARS CONJUNCTION CLASS STOPOVER MISSION MISSION DATA FOR 2042 OPPOSITION

APARTURE PLANET - EARTH

DEPARTURE PLANET - EARTH ARRIUAL PLANET - MARS

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MARS CONJUNCTION CLASS STOPOVER MISSION MISSION DATA FOR 2044 OPPOSITION TABLE 29.

DEPARTURE PLANET - EARTH ABRIVAL PLANET - HARS

DEPARTURE PLANET - EARTH ARRIVAL PLANET - MARS

•.946 1.499 •.15 •.743 84.8 265.4-22.8 2.89 1.013 1.646 2.45 1.104 91.5 15.0 29.3 15.35 •••987 1.499 •••83 •.743 84.8 265.9-23.7 2.89 1.013 1.646 2.45 1.104 91.5 15.0 29.3 15.35 0.987 1.498 -0.09 0.745 84.7 265.2-24.6 2.80 1.013 1.645 2.55 1.104 91.4 15.0 30.2 15.42 0.988 1.498 -0.23 0.745 84.7 265.5-25.8 2.80 1.013 1.645 2.55 1.104 21.4 15.0 30.2 15.42 ••988 1.497 -•.39 •.745 84.7 265.8-27.0 2.89 1.013 1.645 2.55 1.104 91.4 15.0 30.2 15.42 **0.948 1.497 -0.55 0.746 84.6 265.0-28.3 2.81** 1.013 1.644 2.65 1.104 91.4 15.0 31.1 15.52 ••988 1.496 -•.72 •.747 84.4 264.2-29.5 2.82 1.012 1.645 2.61 1.104 91.7 14.4 30.3 15.83 H DO 5.58741 3.95433 1.59728 1.03077 11.92370 5.58387 3.95418 1.59728 1.83077 11.90212 5.58237 3.95381 1.59820 1.03088 11.89254 5.58289 3.95385 1.59820 1.03088 11.89479 5.59465 3.95497 1.59939 1.03100 11.94769 5.52645 3.95404 1.59820 1.03088 11.91320 5.68739 3.95610 1.60332 1.03143 11.99976 ETAO ມ ມີ ມີ ≪ ⊲ œ œ ETA4 S ISA S O ETAB PERIH APNEL I 2 PERIH APNEL I 2 ETAZ ETAL SMA THETT THET2 SMA THETT THET2 1.243 347.0 560.9 1.329 152.9 367.6 1.243 348.8 564.7 1.329 152.9 367.6 1.243 36.5 561.5 1.329 153.6 367.4 1.243 352.4 561.4 1.243 354.3 561.3 1.329 153.6 367.4 1.242 356.0 562.1 1.329 154.3 367.2 1.242 357.8 563.0 1.20916 1.27141 1.20825 1.27230 1.20825 1.27230 1.20825 1.87230 2.13999 1.23966 1.20488 1.27764 1.20916 1.27141 1.20763 1.27338 **T** F II UI PSII ECCEN 92.2 0.206 82.2 0.238 29.3 1.82 1.103 91.9 0.206 -7.0 -3.23 0.710 82.2 0.238 38.4 2.48 1.194 99.4 0.295 -8.6 -3.32 0.706 82.7 0.238 30.9 1.93 1.103 91.6 0.206 -7.7 -3.31 0.709 82.3 0.238 32.8 2.05 1.103 91.3 0.205 -7.7 -3.31 0.709 82.3 0.238 91.0 0.205 82.3 0.238 90.7 0.205 82.5 0.238 1.13025 2.13403 1.23672 1.13425 2.13152 1.23638 1.13355 2.13038 1.23634 1.23789 1.13355 2.13458 1.23641 1.23684 **BBS** 1.13355 2.13252 1.13756 2.13556 912.0 1.73 1.75 917.0 -3.23 0.7-34.8 2.19 1.104 -7.7 -3.31 0.709 36.6 2.33 1.194 -8.4 -3.39 0.708 Ĩ 1.15107 ž DECL 67568.0 67874.0 8.92 162.7 68214.0 68564.0 6.65 120.8 1.7863 2.3882 3.55586 0.99860 0.59044 67564.0 67872.0 9.10 163.9 68212.0 68564.0 5.65 121.2 1.7765 2.3949 3.56793 1.0007 0.89396 67566.0 67872.0 8.08 163.2 68212.0 68564.0 6.69 121.2 1.7766 2.3949 3.56238 0.99876 0.89396 67570.0 67874.0 8.93 161.9 68214.0 68564.0 6.65 120.8 1.7863 2.3882 0.89044 3.56032 0.99888 0.89044 67572.0 67874.0 9.03 151.1 64214.0 68564.0 5.65 120.8 1.7863 2.3882 3.56459 1.00051 0.89044 67574.0 67876.0 9.18 160.6 68216.0 68564.9 6.62 120.4 1.7959 2.3812 3.57130 1.00450 0.88882 17576.0 57878.0 9.39 160.0 18218.0 58565.0 6.51 120.7 « « « « 0.573.0 2 CORD8 C3 2.3740 DEPART ARRIVE RESTART PETURN CORDI COR DVI EUZ 3 1.8453 3.58146

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TABLE 30. PLANETARY MISSION STAGE SIZING FOR MARS CONJUNCTION CLASS MISSION, OPPOSITION 2031 TO 2040

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PLANETARY MISSION STAGE SIZING FOR MARS CONJUNCTION CLASS MISSION, OPPOSITIONS 2042 AND 2044 TABLE 31.

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APPENDIX

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a. SINGLE PLANE TRANSFER GEOMETRY

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b. RELATIONSHIPS IN THE TRANSFER PLANE

Figure 26. Interplanetary transfer geometry.

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DESCRIPTION OF PLANETARY TRAJECTORY DATA

Following is a list of definitions and descriptions for the parameters contained in the tabular data.

SYMBOL DESCRIPTION

- DEPART Julian date of departure; reckoned from JD 240-0000.
- PASS Julian date of passage; reckoned from JD 240-0000.
- ARRIVE Julian date of arrival; reckoned from JD 240-0000.
- C3 Hyperbolic excess velocity squared; expressed in kilometers squared per second squared.
- C3-DD Square of the hyperbolic excess velocity at the departure planet; expressed in kilometers squared per second squared.
- C3-AD Square of the hyperbolic excess velocity at the arrival planet; expressed in kilometers squared per second squared.
- RA Right ascension of hyperbolic excess velocity vector (hyperbolic asymptote); measured in degrees along the local planetary equator eastward from the "vernal equinox," i.e., where in Sun's path rises above the planet's equator.
- DECL Declination of hyperbolic excess velocity vector; measured in degrees positively northward and negatively southward from the planet's equator.
- 11 Inclination of transfer orbit to the planet's orbit at the start of the transfer in degrees; zero is forward, -90 deg is perpendicular southward, +90 deg is perpendicular northward and ±180 deg is backward.
- Vi Heliocentric speed; at the start of the transfer normalized to Earth's mean orbital speed.
- PSI1 Heliocentric angle at the start of the transfer, in degrees; measured counterclockwise from the outward heliocentric radius vector to the velocity vector in the transfer plane.
- ECCEN Eccentricity of the heliocentric transfer conic.
- SMA Semi-major axis of the transfer conic, in AU (asterisk indicates nearparabolic transfer).
- THET1 True anomaly at the start of the transfer, in degrees; reduced to $0 < \theta < 360$ deg.
- THET2 True anomaly at the end of the transfer. in degrees.
- PERIH Perihelion distance of the transfer conic, in AU. This value is listed only if the vehicle traverses perihelion during the transfer.

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SYMBOL DESCRIPTION

- APHEL Aphelion distance of the transfer conic, in AU. This value is listed only if the vehicle traverses aphelion during the transfer.
- V2 Ileliocentric speed at the end of the transfer; normalized to Earth's mean orbit speed.
- PSI2 Heliocentric angle at the end of the transfer, in degrees; measured counterclockwise from the outward heliocentric radius vector to the velocity vector in the transfer plane.
- VHP Hyperbolic excess velocity; expressed in kilometers per second.
- RCP Radius of closest approach to passage planet, in planet radii.
- KAPPA Bend angle at passage planet measured from the positive approach asymptote to the positive departure asymptote written in degrees.
- VP Velocity at pericenter passage; normalized to Earth's mean orbital speed.
- AH Semi-major axis of the passage hyperbola; in planet radii.
- EH Eccentricity of the passage hyperbola.
- ANMAX Maximum value of true anomaly; also the angle between the pericenter radius vector and either asymptote; in degrees.
- INH Inclination of the plane of the passage hyperbola to the equator of the passage planet, in degrees.
- RAP DECP Right ascension and declination of pericenter with respect to the passage planet's equator and equinox; in degrees.
- RAS DECS Right ascension and declination of the Sun with respect to the passage planet's equator and equinox; in degrees.
- ETA Angle between the vector to pericenter and the vector to the Sun, in degrees.

PASS

- CONDITION Lighting condition at sub-pericenter point (Light Side or Dark Side).
- DVCP Power maneuver required at swingby planet, in kilometers per second.
- CORD1 Communication distance between Earth and the first planet encountered, at the time of encounter; in AU.
- CORD2 Communication distance: in AU <u>Direct missions</u>; the distance between the stopover planet and Earth at the time of arrival at the stopover planet. <u>Outbound swingby</u>; the distance between the stopover planet and Earth at the time of arrival at the stopover planet.

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SYMBOL DESCRIPTION

CORD2

- (Cont.) Inbound swingby; the distance between the stopover planet and Earth at the time of departure from the stopover planet.
- CORD 3 Communication distance: in AU <u>Outbound swingby</u>; the distance between the stopover planet and Earth at the time of departure from the stopover planet. <u>Inbound swingby</u>; the distance between the passage planet and Earth at the time of passage.
- DELV1, DV1 Impulsive velocity increment for the Earth departure injection maneuver; in kilometers per second.
- DELV2, DV2 Impulsive velocity increment for the target planet capture maneuver; in kilometers per second.
- DELV3, DV3 Impulsive velocity increment for the target planet departure injection maneuver; in kilometers per second.
- DELV4, DV4 Impulsive velocity increment for the Earth braking maneuver; in kilometers per second.
- GAMA1 Weight ratio for the Earth departure injection maneuver.
- GAMA2 Weight ratio for the target planet capture maneuver.
- GAMA3 Weight ratio for the target planet injection maneuver.
- GAMA4 Weight ratio for the Earth braking maneuver.
- ETA1 Payload ratio for the Earth departure injection maneuver.
- ETA2 Payload ratio for the target planet capture maneuver.
- ETA3 Payload ratio for the target planet injection maneuver.
- ETA4 Payload ratio for the Earth braking maneuvers.
- ETA0 Ratio of terminal weight to the initial weight in Earth orbit.

STS UPPER STAGE SIZING PROGRAM PLA. 'ETARY MISSION STAGE SIZING

DEFINITION OF VARIABLES

- SYMBOL DEFINITION
- W_{PL} Mission module (lb).
- Wmem Mars excursion model (lb).
- WPROBE 1 Planetary probes ejected before braking into Mars orbit (lb).
- WPROBE 2 Planetary probes carried into Mars orbit or Pioneer Mars base for conjunction class missions (lb).
- WSTG 3 Total weight of Earth braking stage (lb).
- WF 3 Propellant required by Earth braking stage (lb).
- WBO 3 Dry weight of Earth braking stage (lb).
- WSTG 2 Total weight of Mars braking and escape stage (lb).
- WF 2 Propellant required by Mars braking and escape stage (lb).
- WBO 2 Dry weight of Mars braking and escape stage (lb).
- WSTG 1 Total weight of Earth escape stage (lb).
- WF1 Propellant required by Earth escape stage (lb).
- WEO 1 Dry weight of Earth escape stage (lb).
- WO Total initial weight required in low Earth orbit to perform the mission (lb).
- C_1, C_2 C_3, C_4 Gravity loss factor for each of the four major planetary maneuvers.
- ^ISP1 Specific impulse of first stage propellant (sec).
- I_{SP2} Specific impulse of second stage propellant (sec).
- I_{SP3} Specific impulse of third stage propellant (sec).
- DV1 Impulsive velocity increment for Earth escape maneuver (km/sec).
- DV0 Impulsive velocity increment for Venus powered swingby maneuver on the outbound leg (km/sec)
- DV_{S1} Inpulse velocity increment for Mars braking maneuver (km/sec)

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SYMBOL DESCRIPTION

 DV_{S2} Impulsive velocity increment for Mars escape maneuver (km/sec). DV_4 Impulsive velocity increment for Earth braking maneuver (km/sec). G_1, G_2, G_3 Equal (1/ λ_I = 1) where λI is the mass fraction of stage 1, 2, and 3.

7)

APPROVAL

4)

MARS EXPLORATION VENUS SWINGBY AND CONJUNCTION CLASS MISSION MODES TIME P.3RIOD 2000 TO 2045

By Archie C. Young, John A. Mulqueen, and James E. Skinner

The information in this report has been reviewed for technical content. Review of any information concerning Department of Defense or nuclear energy activities or programs has been made by the MSFC Security Classification Officer. This report, in its entirety, has been determined to be unclassified.

WILLIAM R. MARSHALL munchas

Director, Program Development

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