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Bhatt

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[54] FIBER REINFORCED CERAMIC MATERIAL
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represented by the Administrator of
the National Aeronautics and Space
Administration, Washington, D.C.
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[22] Filed: Apr. 15, 1987

Related U.S. Application Data

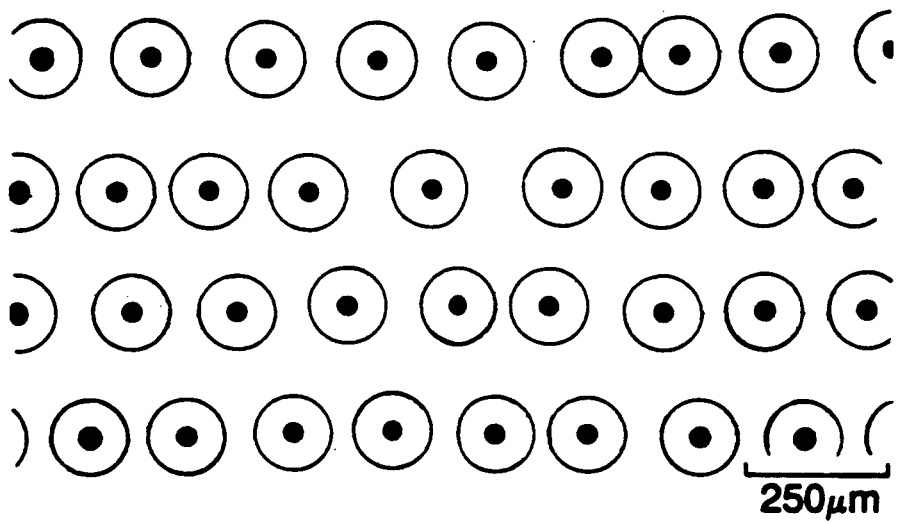
[62] Division of Ser. No. 886,149, Jul. 16, 1986, Pat. No.
4,689,188.
[51] Int. Cl.⁴ B32B 9/00; B32B 19/00;
C04B 35/02
[52] U.S. Cl. 428/698; 428/288;
428/367; 428/375; 428/390; 428/408; 428/446;
501/95
[58] Field of Search 428/408, 367, 446, 375,
428/390, 288, 289; 501/95

References Cited			
U.S. PATENT DOCUMENTS			
3,819,786	6/1974	May	264/63
3,926,656	10/1974	Mangels	106/272
4,004,937	11/1975	Masaki	106/54
4,158,687	6/1979	Yajima et al.	264/60
4,275,095	6/1981	Warren	427/249
4,285,895	8/1981	Mangels et al.	264/65
4,315,968	2/1982	Suplinskas et al.	428/367
4,340,636	7/1982	Debolt et al.	428/367 X
4,472,476	9/1984	Veltri et al.	428/367 X
4,507,354	3/1985	Ishikawa et al.	28/698 X

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[57] ABSTRACT
A strong and tough SiC/RBSN composite material
comprises silicon fibers and a reaction bonded silicon
nitride matrix. This composite material may be used at
elevated temperatures up to at least 1400° C.

5 Claims, 2 Drawing Sheets



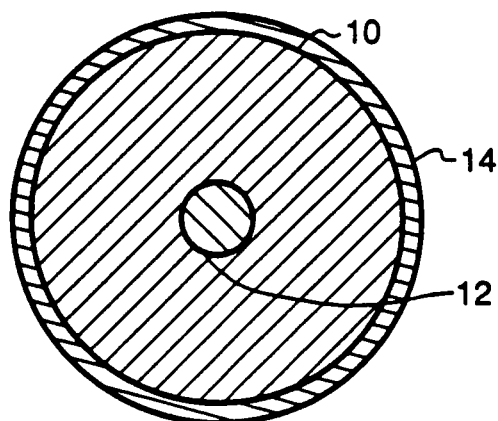


Fig. 1

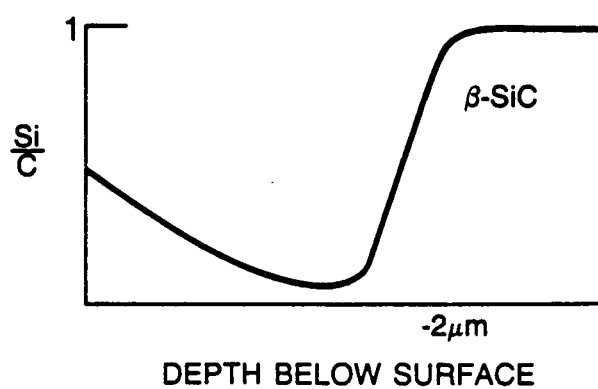


Fig. 2

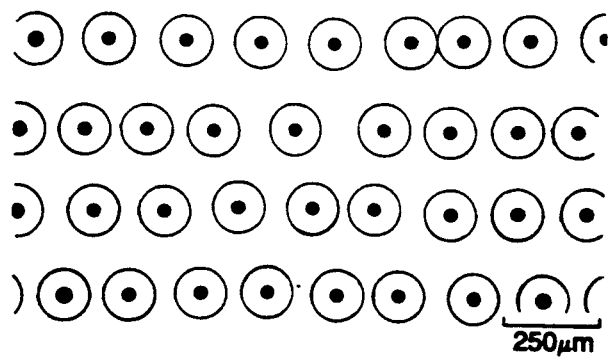


Fig. 3

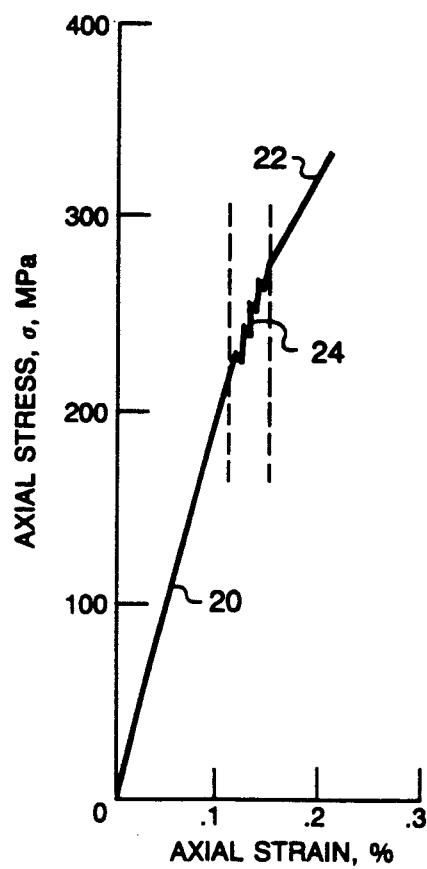


Fig. 4

FIBER REINFORCED CERAMIC MATERIAL

ORIGIN OF THE INVENTION

The invention described herein was made by an employee of the U.S. Government and may be manufactured and used by or for the Government for governmental purposes without the payment of any royalties thereon or therefor.

STATEMENT OF COPENDENCY

This application is a division of application Ser. No. 886,149 which was filed July 16, 1986 and issued as U.S. Pat. No. 4,689,188 or Aug. 25, 1987.

TECHNICAL FIELD

This invention is concerned with an improved fiber reinforced ceramic material. The invention is particularly directed to a reaction-bonded silicon nitride (RBSN) matrix material that is reinforced with fiber.

Because of their lightweight, excellent oxidation resistance, high-temperature strength, environmental stability, and nonstrategic nature, silicon-based ceramics are candidate materials for high performance advanced gas turbine and diesel engines. However, the use of these materials is severely limited because of their inherent flaw sensitivity and brittle behavior. The reinforcement of ceramics by high strength, high modulus, continuous length ceramic fibers should yield stronger and tougher materials. Glass matrix composites reinforced by polymer derived silicon carbide fibers have clearly demonstrated the feasibility of obtaining strong and tough materials. These newly developed composites, however, are presently limited in temperature capability by matrix properties, interfacial reactions, and by thermal instability of the fibers above about 1000° C.

A reaction-bonded silicon nitride (RBSN) material has been suggested for use in advanced heat engine applications. The RBSN material is lightweight, has ease of fabrication, and near net shape capability. However, the use of RBSN material for structural applications is limited because of the low strength of the material as well as its brittle behavior.

In the past preforms for SiC fiber reinforced RBSN composites were prepared by either slip casting or plasma spraying. In the slip casting method, a slurry comprising silicon powder, fugitive polymer binder, and a dispersing medium is poured into a mold provided with a cavity filled with ceramic fibers oriented in the desired directions. The mold is agitated ultrasonically to remove entrapped molecules between fibers and slurry and to distribute the fibers in the slurry. The slurry and the fibers are heated in an oven at a controlled rate to drive off fugitive polymer binders and the dispersing medium. The resulting preform is further densified by conventional ceramic processing methods, such as reaction bonding or sintered reaction bonding.

In the plasma spray method, composite preforms are prepared by spraying molten silicon onto an array of fibers. These preforms are further treated in a nitrogen atmosphere at an elevated temperature to convert the silicon to a silicon nitride matrix.

One of the problems encountered in using these prior art procedures is the lack of control over the distribution and alignment of the fibers in the composite. Also the lack of a suitable surface coating on the fibers caused a fiber/matrix reaction which resulted in poor composite strength and brittle behavior. The use of high

temperature densification methods additionally produced in intrinsic weakening of the fibers.

It is, therefore, an object of the present invention to provide a strong and tough SiC/RBSN composite material for use at temperatures up to 1400° C.

Another object of the invention is to provide a method of making such a composite material by utilizing optimum processing variables and using a SiC fiber having a suitable surface coating.

BACKGROUND ART

Yajima et al U.S. Pat. No. 4,158,687 discloses composite materials that are reinforced with SiC fibers which are produced by imbedding the fibers in, or layering with, a powdery matrix (Si_3N_4), pressing, and sintering. Organosilicon binders may be used, but the matrix is not reaction bonded.

Reaction bonding is disclosed in U.S. Pat. Nos. 4,285,895 to Mangels et al and 3,819,786 to May. In the Mangels et al patent reaction bonded Si_3N_4 is densified by heating under nitrogen gas pressure in the presence of a densification aid which is an oxide of Mg, Y, Ce, or Zr. In the May patent silicon nitride articles are made from a dough-like mixture of Si and binder. The articles are heated in nitrogen after hot-milling and the removal of trichloroethylene from the binder at 130° C.

U.S. Pat. No. 3,926,656 to Mangels is concerned with a multiple mixture containing silicon powder. An injection molding composition of Si powder, paraffin wax, zinc stearate, and Fe_2O_3 is used to form molded parts which are nitrated to form Si_3N_4 articles.

U.S. Pat. No. 4,004,937 to Masaaki discloses the use of nickel oxide as a sintering aide for silicon nitride. The silicon nitride ceramic material contains at least one of MgO, ZnO, and NiO in addition to at least one of Al_2O_3 , Cr_2O_3 , Y_2O_3 , TiO_2 , and SnO_2 . The mixed powders are formed into a green compact and sintered in an inert gas.

DISCLOSURE OF INVENTION

A SiC fiber reinforced reaction bonded silicon nitride matrix composite material (SiC/RBSN) is produced by hot pressing alternate layers of mats of specially coated SiC fibers and silicon monotapes. The resulting silicon carbide/silicon preforms are nitrated at elevated temperatures in N_2 or N_2/H_2 to form a composite material comprising SiC fibers and a reaction bonded silicon nitride matrix. This composite material may be used in advanced engines operating at temperatures above about 1200° C.

DESCRIPTION OF THE DRAWING

The details of the invention will be described in connection with the accompanying drawings in which

FIG. 1 is an enlarged cross section view of a chemical vapor deposited silicon carbide fiber utilized in the composite material of the present invention,

FIG. 2 is a graph showing the composition profile of the carbon rich coating on the surface of the silicon carbide fiber shown in FIG. 1,

FIG. 3 is a photomicrograph showing a typical cross section of a SiC/RBSN composite material produced in accordance with the present invention showing the fiber distribution, and

FIG. 4 is a graph showing the tensile stress-strain behavior for a 20 vol % SiC fiber/RBSN composite

material at room temperature showing linear and non-linear ranges.

BEST MODE FOR CARRYING OUT THE INVENTION

The fiber reinforced reaction bonded silicon nitride matrix material of the present invention utilizes a commercially available silicon powder having a high purity. To reduce particle size and promote reactivity during nitridation, the powder is milled for 7 to 24 hours in an attrition mill utilizing Si_3N_4 balls, a Si_3N_4 container, and a Si_3N_4 arm. A typical milling charge is 297 grams of silicon powder, 3 grams of nickel oxide (NiO), and 1000 grams of heptane solvent. The amount of the nitriding aid, NiO, ranges from one to seven percent by weight. While NiO is the preferred nitriding aid, it is contemplated that other materials, such as Fe_3O_4 , MgO , and Al_2O_3 , may be used for this purpose.

After this milling operation, the silicon powder is dried in an oven to evaporate the milling medium, which is heptane. The dried silicon powder is used for preparing silicon monotapec.

The chemical analysis, average surface area, and particle size of the powder before and after attrition milling are shown in table 1. It is apparent from the table that after the attrition milling there is a significant increase in the oxygen and carbon contents and essentially no increase in the iron content. The surface area of the powder increased from about $1.6 \text{ m}^2/\text{g}$ to about $10 \text{ m}^2/\text{g}$ while the average particle size decreased from $6 \mu\text{m}$ to $0.4 \mu\text{m}$.

TABLE I

Material	Oxygen wt %	Carbon wt %	Nitrogen wt %	Iron wt %	Surface Area, m^2/g	Average Particle Size, μm
As-Received Silicon Powder	0.43	0.025	0.0004	0.60	1.644	6.0
Attrition Milled Silicon Powder	1.20	0.31	0.07	0.60	10.216	0.4

Silicon carbide fibers are produced by chemical vapor deposition (CVD) from methyltrichlorosilane onto a heated carbon monofilament which is drawn continuously through a conventional deposition reactor. Different surface coatings are deposited onto the silicon carbon fibers by introducing hydrocarbon gas or a mixture of hydrocarbon gas or silane vapor near the exit port of the reactor. An enlarged view of the fiber cross section is shown in FIG. 1.

The fiber consists essentially of a silicon carbide sheath 10 surrounding a pyrolytic graphite coated carbon core 12. The silicon carbide sheath 10 has an outer diameter of about $142 \mu\text{m}$ while the graphite coated carbon core 12 has a diameter of about $37 \mu\text{m}$. The silicon carbide sheath 10 is entirely comprised of columnar B-SiC grains growing in a radial direction with a preferred {111} orientation.

The fiber has a surface coating 14 comprising an overlayer with a high silicon/carbon ratio on top of an amorphous carbon layer. The total thickness of the coating 14 is about $2 \mu\text{m}$. The composition profile of the carbon rich coating 14 is shown in FIG. 2. The average room temperature tensile strength of the starting fiber is greater than 3.8 GPa.

A silicon carbide fiber mat is prepared by winding the silicon carbide fiber shown in FIG. 1 on a metal drum at a predetermined spacing. The fiber is then coated with a polymer slurry by using a paint brush or a pressure

spray gun. This polymer slurry comprises a low glass transition temperature polymer and a solvent. The fiber mat is dried and cut to the required dimensions.

The silicon carbide fiber reinforced reaction bonded silicon nitride matrix composite material of the present invention utilizes both the silicon carbide fiber mats and silicon monotapecs. These silicon monotapecs are produced by mixing the previously milled silicon powder, NiO, a fugitive polymer binder, such as a polybutylmethacrylate or polytetrafluoroethylene known commercially as Teflon, and a standard solvent. This material is mixed in a blender for about 15 minutes, and the contents are filtered to remove excess solvent.

The resulting slurry is poured on a hot skillet to drive off the remaining solvent. The resulting polymer dough is rolled to a desired thickness to form the silicon monotapec. This tape is then cut to a predetermined size.

Alternate layers of silicon carbide fiber mat and silicon monotapec are stacked in an open channel molybdenum die. These alternate layers are then hot pressed in a vacuum furnace to form a green compact or preform. The silicon cloth and molybdenum die parts are separated by graphfoil to prevent any reaction between the preform and the die. As shown in Table II, two types of preforms were produced to illustrate the beneficial technical effects of the invention. One preform contained about 23 volume fraction of SiC fibers while the other contained about 40 volume percent. The volume fraction of fibers in each green compact was varied by controlling the fiber spacing or by adjusting the thickness of the silicon monotapec.

TABLE II

Property	23 Vol % SiC/RBSN Composite	40 Vol % SiC/RBSN Composite	Unreinforced RBSN
Density, gm/cc	2.19	2.36	2.53
Matrix Porosity	0.4	0.4	0.2
Elastic Modulus, GPa	165	230	185
4-Point Bend Ultimate Strength, MPa			
As-fabricated	727	868	262
1200° C.	736	—	400
1400° C.	592	—	345
Tensile Ultimate Strength, MPa			
As-fabricated	352	536	—
Tensile Strain, %			
1st Matrix Cracking	0.13	0.13	—
Ultimate Fracture	0.25	0.3	0.14

These preforms are hot pressed in two stages. In the first stage the molybdenum die is heated to about 600° C. in a vacuum furnace at 2° C./min. This material is then maintained at this temperature for about $\frac{1}{2}$ hour to remove the fugitive binder present in the fiber mats and silicon monotapecs.

In the second stage the die temperature is raised to about 1000° C. and pressed at 35 MPa to 138 MPa for about 15 minutes to about one hour. The die is cooled to

room temperature under vacuum conditions. The composite preform is then removed from the die. It is contemplated this hot pressing could be accomplished in another inert environment, such as nitrogen, instead of using a vacuum.

The composite preform is then transferred to a horizontal nitridation furnace comprising a recrystallized Al_2O_3 reaction tube having stainless steel end caps. A nitriding gas, N_2 or $\text{N}_2 + 4\% \text{H}_2$, having a commercial purity is flowed through the furnace before, during, and after nitridation. The nitridation is performed at about 1100°C . to 1400°C . for between 30 to 70 hours. A cross section of the resulting component material is shown in FIG. 3.

Physical and mechanical property data for unidirectionally reinforced SiC/RBSN composite materials fabricated in accordance with present invention and commercially available unreinforced RBSN matrix material at room and elevated temperatures are shown in Table II. Comparison of this data shows that the ultimate fracture strength and strains of the composites were significantly higher than those of unreinforced RBSN matrix material. Also, the composite strengths increased with the increase in fiber volume fraction.

Measurement of room temperature axial tensile strength for 30 volume percent silicon carbide/RBSN composite material after 100 hour exposure at 1200°C . and 1400°C . was 316 MPa and 323 MPa. These values are similar to the value 350 MPa measured for as-fabricated composites. This demonstrated the thermal stability of the composite material.

When the SiC/RBSN composite material is stressed in tension in a direction parallel to the fiber, the composite extends elastically until the RBSN matrix fractures. At this stress level, in contrast with unreinforced RBSN, the composite retains its shape because of fiber bridging of the matrix cracks. The stress-strain behavior for a typical 20 vol % SiC/RBSN composite specimen tested in tension at room temperature is shown in FIG. 4. This test exhibited a first linear range 20 separated from a second linear range 22 by a non-linear range 24.

In addition, because of their high modulus, the SiC fibers bear more load than the matrix they replace. Thus, the composite stress at which the matrix fractures

is greater than that for an unreinforced matrix. On further stressing the composite above the first matrix fracture, the material continues to deform with multiple matrix cracking until the ultimate fracture strength of the fiber is reached. Therefore, the composite is stronger than the unreinforced matrix, and it is tougher as manifested by a high strain to failure and an ultimate non-catastrophic fracture which is fiber controlled. Because of the excellent thermal stability and creep resistance of the CVD SiC fiber, the composite deformation and fracture behavior is temperature independent to 1200°C .

While the preferred embodiment of the invention is disclosed and described it will be apparent that various modifications may be made to the composite material without departing from the spirit of the invention or the scope of the subjoined claims.

I claim:

1. An improved composite material for use at elevated temperatures up to at least 1400°C . comprising a plurality of mats of silicon carbide fibers, each of said fibers comprising
 - a pyrolytic graphite coated carbon core,
 - a silicon carbide sheath covering said coated carbon core,
 - an amorphous carbon layer covering said silicon carbide sheath and an overlayer having a high silicon/carbon ratio covering said amorphous carbon layer, and
- a silicon nitride matrix reaction bonded to said fibers.
2. An improved composite material as claimed in claim 1 wherein the pyrolytic graphite coated carbon core has a diameter of about $37 \mu\text{m}$.
3. An improved composite material as claimed in claim 2 wherein the silicon carbide sheath has a diameter of about $142 \mu\text{m}$.
4. An improved composite material as claimed in claim 1 wherein the silicon carbide fibers comprise about 23 volume fraction of the composite.
5. An improved composite material as claimed in claim 1 wherein the silicon carbide fibers comprise about 40 volume fraction of the composite.

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