

NASA Technical Memorandum 102318  
ICOMP-89-19

# Time Domain Numerical Calculations of Unsteady Vortical Flows About a Flat Plate Airfoil

S.I. Hariharan  
*University of Akron*  
*Akron, Ohio*

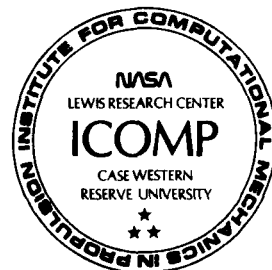
*and Institute for Computational Mechanics in Propulsion*  
*Lewis Research Center*  
*Cleveland, Ohio*

Yu Ping  
*University of Akron*  
*Akron, Ohio*

and

J.R. Scott  
*Lewis Research Center*  
*Cleveland, Ohio*

September 1989



(NASA-TM-102318) TIME DOMAIN NUMERICAL  
CALCULATIONS OF UNSTEADY VORTICAL FLOWS  
ABOUT A FLAT PLATE AIRFOIL (NASA. Lewis  
Research Center) 24 p CSCL 20D

N89-29726

Unclass  
63/34 0234695

# **Time Domain Numerical Calculations of Unsteady Vortical Flows About a Flat Plate Airfoil**

S.I. Hariharan\*

Department of Mathematical Sciences  
University of Akron, Akron, Ohio 44325

and Institute for Computational Mechanics in Propulsion  
Lewis Research Center, Cleveland, Ohio 44135

Yu Ping

Department of Mathematical Sciences  
University of Akron, Akron, Ohio 44325

and

J.R. Scott

National Aeronautics and Space Administration  
Lewis Research Center, Cleveland, Ohio 44135

## **Summary**

A time domain numerical scheme is developed to solve for the unsteady flow about a flat plate airfoil due to imposed upstream, small amplitude, transverse velocity perturbations. The governing equation for the resulting unsteady potential is a homogeneous, constant coefficient, convective wave equation. Accurate solution of the problem requires the development of approximate boundary conditions which correctly model the physics of the unsteady flow in the far field. A uniformly valid far field boundary condition is developed, and numerical results are presented using this condition. The stability of the scheme is discussed, and the stability restriction for the scheme is established as a function of the Mach number. Finally, comparisons are made with the frequency domain calculations by Scott and Atassi, and the relative strengths and weaknesses of each approach are assessed.

---

\*Work funded under Space Act Agreement C99066G.

## 1. Introduction

In recent years, due to the availability of large, high speed computers and improvements in approximation methods, significant progress has been made in computational fluid dynamics in the effort to solve unsteady aerodynamic flow problems. Much of the work that has been done in this area has used the so-called primitive variable approach, wherein a system of governing partial differential equations such as the unsteady Euler or Navier-Stokes equations are solved in time along with certain specified boundary conditions. There are a number of difficulties associated with this approach, including uncertainties about far field boundary conditions, the need for sophisticated algorithms to solve the equations, and the requirements for large computer memory and CPU time.

For many aerodynamic flows of practical interest, the unsteadiness in the flow arises due to small, upstream vortical disturbances (gusts) which are convected downstream and interact with an airfoil or cascade of airfoils. The convected vortical gusts induce unsteady pressure fluctuations on the blade surfaces which can cause forced vibrations and radiate noise into the far field. For this class of "weakly rotational" flows, it is not necessary to solve the full governing equations of fluid motion, but rather a linearized treatment which employs the so-called "rapid distortion" approximation can be used [1]. For potential mean flows, the problem can be formulated in terms of a single, nonconstant-coefficient, inhomogeneous, convective wave equation [2,3].

Among the advantages of the linear approach are the fact that a variety of standard differencing schemes are available for linear problems, the algorithms are simpler, and computational times can be an order of magnitude less than for the primitive variable approach. In addition, it is easier to derive far field boundary conditions which correctly model the physics of the unsteady flow.

Recent articles by Atassi and Scott [4], and Scott and Atassi [5] have described a general numerical scheme which has been developed to implement the linear approach for the purpose of obtaining frequency domain solutions to small amplitude, periodic vortical flows about isolated airfoils. In [4] they show how the motion of a nonuniform flow about a row of moving blades can be modelled as a three-dimensional vortical disturbance convected by the mean flow. They present the general theory for the linearized treatment of such flows, and present unsteady results for symmetric airfoils of arbitrary thickness at various Mach numbers. In [5] they show how the problem simplifies if the thickness of the airfoil goes to zero, and present the details of their frequency domain numerical scheme for the case of thin airfoils in subsonic flows. Comparisons of their numerical results with known analytical solutions to the classical thin airfoil gust response problems indicate that their scheme can accurately predict the unsteady lift on thin airfoils for a large range of Mach numbers and reduced frequencies.

The alternative to the frequency domain formulation reported in [4] and [5] is to solve the

linearized problem in the time domain using a time marching procedure. There are some possible advantages to the time domain approach. For example, in the frequency domain approach, it is necessary to solve a large matrix equation using a direct solver. Due to the time and storage requirements of direct solvers, a time marching scheme could be potentially more storage and computationally efficient. In addition, the frequency domain approach is limited to solving the problem for a single frequency and then superposing solutions for more general disturbances in which multiple frequencies are present. For vortical disturbances that contain a large number of harmonics, many calculations would be necessary and the frequency domain approach would be inefficient. The time domain approach, however, can handle the multiple frequency case in a single calculation, and would in this case be more efficient.

The purpose of the present paper is to present the details of a time domain numerical scheme which has been developed to solve the linearized vortical flow problems as specialized to the case of thin airfoils in a transverse gust. In Section 2 we derive the linearized equations and formulate the boundary value problem for the case of flat plate airfoils. In Section 3 the far field boundary condition is derived, and in Section 4 the basic numerical scheme is presented. Section 5 discusses the details of the calculation of the unsteady response function, and in Section 6 we discuss computed numerical results, and compare with the results of Scott and Atassi as reported in [5].

## 2. Formulation of the Problem

### 2.1 Derivation of Governing Equation

The formulation that is presented here follows that given in the recent review paper of Atassi [1]. Consider the equations of fluid motion for an ideal gas which is inviscid and nonheat conducting. The governing continuity, momentum, and energy equations may be written

$$\rho_t + \nabla \cdot (\rho \vec{v}) = 0 \quad (1)$$

$$\vec{v}_t + (\vec{v} \cdot \nabla) \vec{v} + \frac{1}{\rho} \nabla p = 0 \quad (2)$$

$$s_t + \vec{v} \cdot \nabla s = 0 \quad (3)$$

where  $\rho$ ,  $\vec{v}$ ,  $p$ ,  $s$  are the fluid density, velocity, pressure and entropy respectively.

This is a system of highly nonlinear equations. Since we assume that the upstream vortical disturbances are small compared to the mean velocity, we linearize about the mean flow state and introduce perturbation quantities as follows. Let

$$\rho = \rho_0 + \epsilon \tilde{\rho} \quad (4)$$

$$\vec{v} = \vec{v}_0 + \epsilon \tilde{\vec{v}} \quad (5)$$

$$p = p_0 + \epsilon \tilde{p} \quad (6)$$

$$s = s_0 + \epsilon \tilde{s} \quad (7)$$

where  $\epsilon$  is small compared to the corresponding zeroth order quantities that govern the mean flow whose density, velocity, pressure, and entropy are  $\rho_0$ ,  $\bar{v}_0$ ,  $p_0$ , and  $s_0$ , respectively.

In the case of thin airfoils at zero mean incidence, the mean velocity is a uniform parallel flow in the  $x$  direction, so that  $\bar{v}_0 = v_0 \bar{e}_x$ . The first order asymptotic expansion ( $O(\epsilon)$ ) is thus

$$\frac{1}{\rho_0 c_0^2} \left( \frac{\partial}{\partial t} + v_0 \frac{\partial}{\partial x} \right) \tilde{p} = -\nabla \cdot \bar{u} \quad (8)$$

$$\rho_0 \left( \frac{\partial}{\partial t} + v_0 \frac{\partial}{\partial x} \right) \bar{u} = -\nabla \tilde{p} \quad (9)$$

$$\left( \frac{\partial}{\partial t} + v_0 \frac{\partial}{\partial x} \right) \tilde{s} = 0, \quad (10)$$

where  $c_0 = \sqrt{\gamma p_0 / \rho_0}$  is the acoustic speed.

Following the traditional splitting of the unsteady velocity into solenoidal and irrotational components [1], let

$$\bar{u} = \bar{u}_1 + \nabla \phi \quad (11)$$

where  $\bar{u}_1$  satisfies

$$\nabla \cdot \bar{u}_1 = 0 \quad (12)$$

and

$$\left( \frac{\partial}{\partial t} + v_0 \frac{\partial}{\partial x} \right) \bar{u}_1 = 0. \quad (13)$$

Using equations (11) and (13) in (9), there results

$$\rho_0 \left( \frac{\partial}{\partial t} + v_0 \frac{\partial}{\partial x} \right) \nabla \phi = -\nabla \tilde{p} \quad (14)$$

or

$$\rho_0 \left( \frac{\partial}{\partial t} + v_0 \frac{\partial}{\partial x} \right) \phi = -\tilde{p} + f(t),$$

where  $f(t)$  is an arbitrary function of  $t$ . At upstream infinity  $\phi$ ,  $\phi_x$  and  $\tilde{p}$  vanish. From this we deduce that

$$\rho_0 \left( \frac{\partial}{\partial t} + v_0 \frac{\partial}{\partial x} \right) \phi = -\tilde{p}. \quad (15)$$

Substituting (15) into (8) and using (11) one gets

$$\left( \frac{\partial}{\partial t} + v_0 \frac{\partial}{\partial x} \right)^2 \phi = c_0^2 \nabla^2 \phi. \quad (16)$$

Equation (16) is the governing equation for the unsteady potential for the problem of small disturbances to a uniform flow past a flat plate airfoil.

The total velocity field  $\vec{v}$  is given by

$$\vec{v} = v_0 \vec{e}_x + \vec{u}_1 + \nabla \phi. \quad (17)$$

We observe that once the potential field is determined, the pressure and the velocity are determined from equations (15) and (17), respectively.

## 2.2 Boundary Conditions

Now consider the problem of flow past a flat plate airfoil at zero degrees mean incidence with small amplitude, unsteady velocity disturbances imposed upstream (See Figure 1). The mean flow is a uniform parallel flow in the  $x$  direction, and the  $y$  direction is taken to be perpendicular to the airfoil. Now from conditions (12) and (13) we conclude that  $\vec{u}_1$  must be of the form

$$\vec{u}_1 = \vec{u}_1(x - v_0 t, y) \quad (18)$$

and that the upstream disturbances are convected by the mean flow. Here

$$\vec{u}_1 = w \vec{e}_x + v \vec{e}_y, \quad (19)$$

where  $w$  and  $v$  must be chosen to satisfy the divergence free condition (12). The unsteady velocity  $\vec{u}$  must satisfy  $\vec{u} \rightarrow \vec{u}_1$  as  $x \rightarrow -\infty$ . Therefore  $\phi$  must satisfy

$$\nabla \phi \rightarrow 0 \quad \text{as} \quad x \rightarrow -\infty. \quad (20)$$

At the airfoil surface, the normal component of the velocity must vanish. This leads to the requirement that

$$v + \frac{\partial}{\partial y} \phi = 0. \quad (21)$$

In the wake behind the airfoil  $\phi$  is not continuous but must satisfy a jump condition determined by the continuity of the unsteady pressure. Applying (15) on each side of the vortex sheet behind the airfoil leads to the condition

$$\left( \frac{\partial}{\partial t} + v_0 \frac{\partial}{\partial x} \right) \Delta \phi = 0, \quad (22)$$

where  $\Delta \phi$  is the jump in  $\phi$  across the wake.

## 2.3 Nondimensionalization of the Problem

For computational purposes it is convenient to nondimensionalize the problem. We therefore introduce the following normalization. Let

$$T = \frac{v_0}{l/2} t \quad (23)$$

$$X = \frac{x}{l/2} \quad (24)$$

$$Y = \frac{y}{l/2} \quad (25)$$

$$c = \frac{c_0}{v_0} \quad (26)$$

In the above nondimensionalization,  $l$  is the chord length of the airfoil. Denoting  $T, X, Y$  by  $t, x$  and  $y$  again respectively, the problem takes the following form:

$$\frac{\partial^2 \phi}{\partial t^2} + 2 \frac{\partial^2 \phi}{\partial t \partial x} + \frac{\partial^2 \phi}{\partial x^2} = c^2 \nabla^2 \phi \quad \text{governing equation} \quad (27)$$

$$\frac{\partial}{\partial t} \Delta \phi + \frac{\partial}{\partial x} \Delta \phi = 0 \quad \text{in the wake} \quad (28)$$

$$v + \frac{\partial}{\partial y} \phi = 0 \quad \text{airfoil surface} \quad (29)$$

$$\nabla \phi \rightarrow 0 \quad \text{as } x \rightarrow -\infty. \quad (30)$$

Note that the quantity  $c$  is now the inverse Mach number of the mean flow.

### 3. Absorbing Boundary Conditions

The boundary value problem prescribed in equations (27)-(30) is an open domain problem. In order to obtain numerical solutions to this problem it is necessary to truncate the domain onto a finite region as shown in Figure 2. Because we consider only flat plate airfoils with zero thickness, the solution  $\phi$  is an odd function with respect to  $y$ . Therefore it is only necessary to solve the problem in the upper half plane. The computational domain is the rectangular region defined by  $[-L, L] \times [0, H]$ .

Because the open domain problem has been truncated onto a finite region with fixed boundaries, and since the gust response problem is essentially a wave propagation problem, it is necessary to derive far field boundary conditions which allow outgoing acoustic waves to pass through the far field boundary without being reflected back into the computational domain. Since the governing equation is a convective wave equation, and not the regular wave equation, there are difficulties in deriving nonreflecting far field boundary conditions. We shall present an approach due to [8] which is used in our computations, and we will now describe this approach in detail.

Generalization of the ideas to follow are found in Hariharan and Hagstrom [8]. This procedure has similarities to that of Bayliss and Turkel [6], but is obtained through a systematic derivation and has general validity. The condition which is developed in [6] is valid only in the outflow, whereas the conditions developed in [8] are valid uniformly in all directions.

To present the ideas, we consider the convective wave equation in its original form

$$\phi_{tt} + 2v_0 \phi_{xt} + v_0^2 \phi_{xx} = c_0^2 \nabla^2 \phi. \quad (31)$$

The idea has both a geometrical and an algebraic point of view. The geometrical view is that we seek a transformation in which the waves at infinity are cylindrical with respect to a moving source convected along the  $x$  axis with a speed  $U_\infty$ . We shall now describe this process.

Suppose we consider a wave front at a time  $t = t^*$ . From time  $t = 0$  to this time a source in the flow is convected a distance  $v_0 t^*$ . The signals would have propagated to a distance  $c_0 t^*$ . Thus the first step in deriving the transformation we seek is to develop a relation between  $v_0 t^*$  and  $c_0 t^*$  with respect to a fixed observer. This is depicted in Figure 3.

In this figure, if we label the distance from the origin to the wave front by  $\hat{R} c_0 t^*$ , we obtain

$$c_0^2 \hat{R}^2 + v_0^2 - 2 \hat{R} v_0 \cos \theta = c_0^2 \quad (32)$$

Solving for  $\hat{R}$  we obtain:

$$\hat{R}(\theta) = M_\infty \cos \theta \pm \sqrt{1 - M_\infty^2 \sin^2 \theta} \quad (33)$$

where  $M_\infty = \frac{v_0}{c_0}$ . We conclude on the basis of the behavior near  $\theta = 0$  and  $\theta = \pi$  that only the positive sign is valid.

Thus the desired transformation in which the cylindrical properties are obtained is

$$\hat{r} = \frac{r}{c_0 \hat{R}(\theta)}, \quad \hat{\theta} = \theta. \quad (34)$$

The interpretation is that at  $r = c_0 \hat{R}(\theta) t^*$ ,  $\hat{r} = t^* = \text{constant}$ , the wave front is cylindrical. The transformation appropriate to the scaled equation

$$\phi_{tt} + 2\phi_{xt} + \phi_{xx} = c^2 \nabla^2 \phi \quad (35)$$

is

$$\hat{r} = r S(\theta) \quad (36)$$

where

$$S(\theta) = \frac{1}{\cos \theta + \sqrt{c^2 - \sin^2 \theta}} \quad (37)$$

Now in the new coordinate system  $(\hat{r}, \hat{\theta})$ , which we call the wave front coordinate system, one can represent the solution in the far field following Friedlander [7] in an asymptotic form. That is

$$\phi \sim \frac{f(t - \hat{r})}{\hat{r}^{1/2}} \quad (38)$$

From this, we obtain a radiation boundary condition that is similar to the one that is reported in [1], which is

$$\phi_t + \phi_{\hat{r}} + \frac{\phi}{2\hat{r}} = 0 \quad \text{to } O(\hat{r}^{-\frac{5}{2}}). \quad (39)$$



Thus the desired boundary condition (which we call the H-H condition) is a translation of (39) in the original coordinate system. This results in the boundary condition

$$\phi_t + \frac{1}{S(\theta)}\phi_r + \frac{\phi}{2rS(\theta)} = 0. \quad (40)$$

This condition is implemented in the results reported in the present paper. In the numerical calculations this is implemented as

$$\phi_t + \frac{1}{S(\theta)}[\cos(\theta)\phi_x + \sin(\theta)\phi_y] + \frac{\phi}{2rS(\theta)} = 0. \quad (41)$$

We remark here that, in contrast to the geometric derivation given above, the transformation (37) can also be obtained algebraically by solving the Eikonal equation corresponding to (35) which is

$$1 - 2\sigma_x + \sigma_x^2 = c^2(\sigma_x^2 + \sigma_y^2). \quad (42)$$

#### 4. Numerical Scheme and Stability

##### 4.1 Basic Scheme and the Stability Constraint

The indices used in our numerical calculations are  $i$  and  $j$ , where  $i$  varies from 0 to  $M$  along the  $x$  axis and  $j$  varies from 0 to  $N$  along the  $y$  axis. The index  $n$  is used to denote values at the time  $n\Delta t$ , where  $\Delta t$  is the time step. The grid spacing in the  $x$  and  $y$  directions is denoted by  $\Delta x$  and  $\Delta y$ , respectively.

Unlike the regular wave equation, the convective wave equation yields difficulties in obtaining accurate schemes due to the presence of the mixed derivative term  $\phi_{xt}$ . A scheme that is second order accurate in space and in time was obtained using the following approximations:

$$(\phi_{tt})_{i,j}^n = \frac{\phi_{i,j}^{n+1} - 2\phi_{i,j}^n + \phi_{i,j}^{n-1}}{\Delta t^2} \quad (43)$$

$$(\phi_{tx})_{i,j}^n = \frac{3\phi_{i,j}^{n+1} - 4\phi_{i-1,j}^{n+1} + \phi_{i-2,j}^{n+1} - 3\phi_{i,j}^{n-1} + 4\phi_{i-1,j}^{n-1} - \phi_{i-2,j}^{n-1}}{4\Delta t\Delta x} \quad (44)$$

$$(\phi_{xx})_{i,j}^n = \frac{\phi_{i+1,j}^n - 2\phi_{i,j}^n + \phi_{i-1,j}^n}{\Delta x^2} \quad (45)$$

$$(\phi_{yy})_{i,j}^n = \frac{\phi_{i,j+1}^n - 2\phi_{i,j}^n + \phi_{i,j-1}^n}{\Delta y^2} \quad (46)$$

Note that all the second order derivatives are evaluated using the standard central difference formula except the mixed derivative term. The central difference formula cannot be used for the mixed derivative term, as it results in an unstable scheme. The truncation error corresponding to this

term can be shown to be equal to

$$\begin{aligned} ERROR_{truncation} &= -\frac{\Delta t^2}{12}\phi_{tttt} - \frac{\Delta t^2}{3}\phi_{tttz} \\ &+ \frac{2\Delta x^2}{3}\phi_{tzzz} + \frac{\Delta x^2}{12}(c^2 - 1)\phi_{zzzz} + \frac{\Delta y^2}{12}c^2\phi_{yyyy} \\ &= O(\Delta t^2, \Delta x^2). \end{aligned} \quad (47)$$

Substituting equations (43) - (46) into (27), we obtain the following numerical scheme:

$$\begin{aligned} &\phi_{i,j}^{n+1} - 2\phi_{i,j}^n + \phi_{i,j}^{n-1} \\ &+ 0.5R(3\phi_{i,j}^{n+1} - 4\phi_{i-1,j}^{n+1} + \phi_{i-2,j}^{n+1} - 3\phi_{i,j}^{n-1} + 4\phi_{i-1,j}^{n-1} - \phi_{i-2,j}^{n-1}) \\ &+ (1 - c^2)R^2(\phi_{i+1,j}^n - 2\phi_{i,j}^n + \phi_{i-1,j}^n) - c^2R^2(\phi_{i,j+1}^n - 2\phi_{i,j}^n + \phi_{i,j-1}^n) = 0, \end{aligned} \quad (48)$$

where

$$R = \frac{\Delta t}{\Delta x} = \frac{\Delta t}{\Delta y}.$$

To discuss the stability of the scheme we let the error term be the form

$$E_{i,j}^t = A\xi^i e^{i\alpha\Delta x i} e^{i\beta\Delta y j}. \quad (49)$$

Substituting into (48), we get the error equation

$$\begin{aligned} &\xi^2(1 + 1.5R - 2Re^{-i\alpha\Delta x} + 0.5Re^{-i2\alpha\Delta x}) \\ &+ \xi[-2 + 4(c^2 - 1)R^2 \sin^2(\frac{\alpha\Delta x}{2}) + 4c^2R^2 \sin^2(\frac{\beta\Delta y}{2})] \\ &+ (1 - 1.5R + 2Re^{-i\alpha\Delta x} - 0.5Re^{-i2\alpha\Delta x}) = 0. \end{aligned} \quad (50)$$

We solve this equation numerically for all possible  $\alpha\Delta x$  and  $\beta\Delta y$  for fixed  $R$  and  $c$ , and find that the roots of the equation are all within the unit circle if the following stability criteria are satisfied:

for  $M_\infty = 0.8$ ,  $R < 0.15$ ,

for  $M_\infty = 0.5$ ,  $R < 0.12$ .

Note that there are fictitious points, namely those points at  $j = -1$ ,  $j = N + 1$ ,  $i = -2, -1$ , and  $i = M + 1$ . Thus, the scheme given by equation (48) is valid only in the interior domain, and special treatment is required on the grid line corresponding to  $i = 1$  near the upstream boundary ( $i = 0$ ).

## 4.2 Numerical Differencing Scheme

The finite difference equations used to model the governing equation and various boundary conditions are presented below.

### 4.2a Upstream Boundary

The calculation is begun at the left boundary, where the boundary condition is

$$\phi_t + \frac{1}{S(\theta)} [\cos \theta \phi_x + \sin \theta \phi_y] + \frac{\phi}{2rS(\theta)} = 0. \quad (51)$$

The differencing used at the left boundary is

$$(\phi_t)_{0,j}^n = \frac{1}{\Delta t} (\phi_{0,j}^{n+1} - \phi_{0,j}^n) \quad (52)$$

$$(\phi_x)_{0,j}^n = \frac{1}{2\Delta x} (-3\phi_{0,j}^n + 4\phi_{1,j}^n - \phi_{2,j}^n) \quad (53)$$

$$(\phi_y)_{0,j}^n = \frac{1}{2\Delta y} (\phi_{0,j+1}^n - \phi_{0,j-1}^n), \quad (54)$$

where

$$(\phi)_{0,j}^n = \frac{1}{2} (\phi_{0,j}^{n+1} + \phi_{0,j}^{n-1}) \quad (55)$$

$$(r)_{0,j} = \sqrt{x_0^2 + y_j^2} \quad (56)$$

$$(\cos(\theta))_{0,j} = \frac{x_0}{r_{0,j}} \quad (57)$$

$$(\sin(\theta))_{0,j} = \frac{y_j}{r_{0,j}} \quad (58)$$

$$\frac{1}{S(\theta)} = \frac{1}{S_{0,j}} = \frac{x_0}{r_{0,j}} + \sqrt{c^2 - y_j^2 / r_{0,j}^2}. \quad (59)$$

The finite difference equation for the boundary  $i = 0$  is then given by

$$\begin{aligned} \phi_{0,j}^{n+1} = & \frac{1}{1 + \frac{R}{4rS_{0,j}}} \{ \phi_{0,j}^n \\ & - \frac{R}{2r_{0,j}S_{0,j}} [ x_0(-3\phi_{0,j}^n + 4\phi_{1,j}^n - \phi_{2,j}^n) \\ & + y_j(\phi_{0,j+1}^n - \phi_{0,j-1}^n) + \frac{1}{2}\phi_{0,j}^{n-1} ] \} \quad j = 1 \text{ to } N-1. \end{aligned} \quad (60)$$

### 4.2b Interior Points

For interior points the governing equation is

$$\phi_{tt} + 2\phi_{xt} + \phi_{xx} = c^2 \nabla^2 \phi. \quad (61)$$

We use an  $O(\Delta t^2, \Delta x)$  scheme for those points with  $i = 1$ , for  $j = 1$  to  $j = N$ . The difference approximations are

$$(\phi_{tt})_{1,j}^n = \frac{1}{\Delta t^2} (\phi_{1,j}^{n+1} - 2\phi_{1,j}^n + \phi_{1,j}^{n-1}) \quad (62)$$

$$(\phi_{xx})_{1,j}^n = \frac{1}{\Delta x^2} (\phi_{2,j}^n - 2\phi_{1,j}^n + \phi_{0,j}^n) \quad (63)$$

$$(\phi_{yy})_{1,j}^n = \frac{1}{\Delta y^2} (\phi_{1,j+1}^n - 2\phi_{1,j}^n + \phi_{1,j-1}^n) \quad (64)$$

$$(\phi_{tx})_{1,j}^n = \frac{1}{2\Delta t \Delta x} (\phi_{1,j}^{n+1} - \phi_{0,j}^{n+1} - \phi_{1,j}^{n-1} + \phi_{0,j}^{n-1}) \quad (65)$$

Therefore the difference equation for  $i = 1$ ,  $j = 1$  to  $j = N - 1$  is

$$\begin{aligned} \phi_{1,j}^{n+1} = \frac{1}{1+R} \{ & 2\phi_{1,j}^n - \phi_{1,j}^{n-1} - R(-\phi_{0,j}^{n+1} - \phi_{1,j}^{n-1} + \phi_{0,j}^{n-1}) \\ & + (c^2 - 1)R^2(\phi_{2,j}^n - 2\phi_{1,j}^n + \phi_{0,j}^n) \\ & + c^2 R^2(\phi_{1,j+1}^n - 2\phi_{1,j}^n + \phi_{1,j-1}^n) \} \end{aligned} \quad (66)$$

For interior points where  $i = 2$  to  $i = M - 1$ ,  $j = 1$  to  $j = N - 1$ , the  $O(\Delta t^2, \Delta x^2)$  finite difference equation corresponding to equation (59) is

$$\begin{aligned} \phi_{i,j}^{n+1} = \frac{1}{1+1.5R} [ & 2\phi_{i,j}^n - \phi_{i,j}^{n-1} \\ & - \frac{R}{2}(-4\phi_{i-1,j}^{n+1} + \phi_{i-2,j}^{n+1} - 3\phi_{i,j}^{n-1} + 4\phi_{i-1,j}^{n-1} - \phi_{i-2,j}^{n-1}) \\ & + (c^2 - 1)R^2(\phi_{i+1,j}^n - 2\phi_{i,j}^n + \phi_{i-1,j}^n) \\ & + c^2 R^2(\phi_{i,j+1}^n - 2\phi_{i,j}^n + \phi_{i,j-1}^n) ] \end{aligned} \quad (67)$$

#### 4.2c Airfoil Surface

At the surface of the airfoil the boundary condition is

$$v + \frac{\partial \phi}{\partial y} = 0. \quad (68)$$

The corresponding difference equation is

$$\phi_{i,0}^{n+1} = \frac{1}{3} ( 4\phi_{i,1}^{n+1} - \phi_{i,2}^{n+1} + 2v_{i,0}\Delta y ), \quad i = M_1 \text{ to } M_2, j = 0, \quad (69)$$

where  $M_1$  is the  $i$  index of the grid point at the airfoil leading edge and where  $M_2$  is the  $i$  index of the grid point at the trailing edge.

#### 4.2d Top Boundary

On the top boundary the far field boundary condition is

$$\phi_t + \frac{1}{rS(\theta)} (x_i \phi_x + y_{N-1} \phi_y) + \frac{\phi}{2rS(\theta)} = 0 \quad (70)$$

where the difference approximations are

$$(\phi_t)_{i,N-1}^n = \frac{1}{4\Delta t} (\phi_{i,N}^{n+1} - \phi_{i,N}^{n-1} + \phi_{i,N-2}^{n+1} - \phi_{i,N-2}^{n-1}) \quad (71)$$

$$(\phi_x)_{i,N-1}^n = \frac{1}{2\Delta x} (\phi_{i+1,N-1}^n - \phi_{i-1,N-1}^n) \quad (72)$$

$$(\phi_y)_{i,N-1}^n = \frac{1}{4\Delta y} (\phi_{i,N}^{n+1} - \phi_{i,N-2}^{n+1} + \phi_{i,N}^{n-1} - \phi_{i,N-2}^{n-1}) \quad (73)$$

and where

$$(\phi)_{i,N-1}^n = \frac{1}{2} (\phi_{i,N}^{n+1} + \phi_{i,N-2}^{n-1}) \quad (74)$$

$$(r)_{i,N-1} = \sqrt{x_i^2 + y_{N-1}^2}. \quad (75)$$

The difference equation is then

$$\begin{aligned} \phi_{i,N}^{n+1} = & \frac{1}{1 + \frac{R y_{N-1}}{r_{i,N-1} S_{i,N-1}} + \frac{\Delta t}{4 r_{i,N-1} S_{i,N-1}}} \{ \phi_{i,N}^{n-1} + \phi_{i,N-2}^{n+1} - \phi_{i,N-2}^{n-1} \\ & - \frac{R}{r_{i,N-1} S_{i,N-1}} [2x_i (\phi_{i+1,N-1}^n - \phi_{i-1,N-1}^n) \\ & + y_{N-1} (-\phi_{i,N-2}^{n+1} + \phi_{i,N}^{n-1} - \phi_{i,N-2}^{n-1}) + \frac{\Delta x}{2} \phi_{i,N-2}^{n-1}] \} \\ & \text{for } i = 1 \text{ to } M-1, j = N. \end{aligned} \quad (76)$$

#### 4.2e Downstream Boundary

At the downstream boundary the far field condition is

$$\phi_t + \frac{1}{rS(\theta)} (x_{M-1} \phi_x + y_j \phi_y) + \frac{\phi}{2rS(\theta)} = 0 \quad (77)$$

where the differencing used is

$$(\phi_t)_{M-1,j}^n = \frac{1}{4\Delta t} (\phi_{M,j}^{n+1} - \phi_{M,j}^{n-1} + \phi_{M-2,j}^{n+1} - \phi_{M-2,j}^{n-1}) \quad (78)$$

$$(\phi_x)_{M-1,j}^n = \frac{1}{4\Delta x} (\phi_{M,j}^{n+1} - \phi_{M-2,j}^{n+1} + \phi_{M,j}^{n-1} - \phi_{M-2,j}^{n-1}) \quad (79)$$

$$(\phi_y)_{M-1,j}^n = \frac{1}{2\Delta y} (\phi_{M-1,j+1}^n - \phi_{M-1,j-1}^n) \quad (80)$$

and where

$$(\phi)_{M-1,j}^n = \frac{1}{2} (\phi_{M-1,j}^{n+1} + \phi_{M-1,j}^{n-1}) \quad (81)$$

$$(r)_{M-1,j} = \sqrt{x_{M-1}^2 + y_j^2} \quad (82)$$

The finite difference approximation is then

$$\begin{aligned} \phi_{M,j}^{n+1} = & \frac{1}{1 + \frac{Rx_{M-1}}{r_{M-1,j}S_{M-1,j}} + \frac{\Delta t}{4r_{M-1,j}S_{M-1,j}}} \{ \phi_{M,j}^{n-1} - \phi_{M-2,j}^{n+1} + \phi_{M-2,j}^{n-1} \\ & - \frac{R}{r_{M-1,j}S_{M-1,j}} [x_{M-1}(-\phi_{M-2,j}^{n+1} + \phi_{M,j}^{n-1} - \phi_{M-2,j}^{n-1}) \\ & + 2y_j(\phi_{M-1,j+1}^n - \phi_{M-1,j-1}^n) + \frac{\Delta x}{2}\phi_{M-2,j}^{n-1}] \} \\ & \text{for } i = M, j = 1 \text{ to } N-1. \end{aligned} \quad (83)$$

#### 4.2f Wake Boundary and Corner Points

For the wake boundary condition, we make use of the fact that  $\phi$  is an odd function of  $y$ , so that  $\Delta\phi = \phi^+ - \phi^- = 2\phi^+$ , where  $+$  and  $-$  denote quantities above and below the wake, respectively. The wake boundary condition (28) then becomes

$$\frac{\partial}{\partial t}\phi^+ + \frac{\partial}{\partial x}\phi^+ = 0, \quad (84)$$

and the difference equation is

$$\phi_{i,0}^{n+1} = \phi_{i,0}^{n-1} - R(\phi_{i+1,0}^n - \phi_{i-1,0}^n) \quad \text{for } i = M_2 \text{ to } M-1, j = 0. \quad (85)$$

For the last grid point in the wake ( $i = M, j = 0$ ), we use

$$\phi_{M,0}^{n+1} = \frac{1}{1+R} [ \phi_{M,0}^n + \phi_{M-1,0}^n - \phi_{M-1,0}^{n+1} + R(\phi_{M-1,0}^{n+1} + \phi_{M-1,0}^n - \phi_{M,0}^n) ] \quad (86)$$

Lastly, for the corner points  $\phi_{M,N}$ ,  $\phi_{0,N}$  we just take  $\phi$  to be the average value of the two nearby points, i.e.,

$$\phi_{0,N}^{n+1} = 0.5(\phi_{0,N-1}^{n+1} + \phi_{1,N}^{n+1}) \quad (87)$$

$$\phi_{M,N}^{n+1} = 0.5(\phi_{M,N-1}^{n+1} + \phi_{M-1,N}^{n+1}). \quad (88)$$

## 5. Calculation of the Unsteady Response Function

For purposes of comparing numerical results with known solutions to the classical thin airfoil, gust response problems, we introduce (following Sears [9]) the unsteady response function

$$R(k_1, M_\infty) = \frac{L}{\pi \rho l v_0 e^{(i\nu t)}}, \quad (89)$$

where  $L$  is the unsteady lift, and  $k_1 = \frac{\nu l}{2v_0}$ , where  $\nu$  is the angular frequency of the incident disturbance, is the reduced frequency. The lift is calculated by integrating the unsteady pressure over the surface of the airfoil.

Since we solve the problem in the time domain, and not the frequency domain, we obtain the response function through the use of the Fourier transform. The upstream velocity disturbance is taken to be a wide band Gaussian pulse, and we obtain the airfoil response as a function of the reduced frequency as explained below.

Suppose that the value of the disturbance  $v(x, y, t)$  at the center of the airfoil is  $f(t)$ . Then the time spectrum of this disturbance will be

$$F(\omega) = \frac{1}{\sqrt{2\pi}} \int_{-\infty}^{+\infty} f(t) e^{(i\omega t)} dt. \quad (90)$$

The response of the airfoil to the disturbance will be

$$g(t) = \frac{L(t)}{\pi \rho c v_0}, \quad (91)$$

where the lift  $L(t)$  is

$$L(t) = \int_{\Gamma} \tilde{p}(t) ds. \quad (92)$$

By taking the Fourier transformation of  $g(t)$ , we get the representation of  $g(t)$  in the frequency domain  $G(\omega)$ , and the frequency response function of the airfoil is

$$R(\omega) = \frac{G(\omega)}{F(\omega)}. \quad (93)$$

For example, let us consider the case where

$$f(t) = \sqrt{2a} e^{(-at^2)}. \quad (94)$$

The spectrum of  $f(t)$  is then

$$F(\omega) = e^{(-\frac{\omega^2}{4a})}, \quad (95)$$

and the pulse function (disturbance) for the problem under consideration at time  $t$  and at a point  $x$  is  $f(t - x)$ .

The numerical integration involved in evaluating the transforms  $F(\omega)$  and  $G(\omega)$  depends on the value of the amplitude  $a$ . For example, for a value of  $a = 4$ , Figure 4 shows the behavior of  $f(t)$ . Since the decay rate of this function is fast, one may choose a time interval of the form  $[-T_0, T_0]$ . A typical value of  $T_0$  for our grids was 6.975. The corresponding transformation is given in Figure 5. For numerical purposes, the index for the time begins at  $t = -T_0$  so that  $n = 0$  corresponds to  $t = -T_0$ . At the time level  $n=4000$ , the pulse has passed the airfoil and the response function of  $g(t)$  tends to zero. Therefore this is a good place to truncate the response function  $g(t)$  numerically. We can then calculate its Fourier transformation  $G(\omega)$  very easily using Riemann sums. The response function  $g(t)$  and its transforms are given in Figures 6 and 7, respectively. Finally, using (93), we get the frequency response function  $R(\omega)$ . The various frequency response functions given in Figures 8 – 13 are obtained by plotting  $Re[R(\omega)]$  vs  $Im[R(\omega)]$ .

## 6. Results and Discussion

### 6.1 Comparison of Numerical Results with Known Solutions

In order to assess the accuracy of the numerical scheme that has been developed, we compare our computed results with known solutions to the classical thin airfoil gust response problems. Because the stability constraint of our scheme depends on the Mach number, we are limited to calculating cases for which the Mach number is not small. For comparison purposes we present results for Mach numbers of .5 and .8.

Figure 8 shows a comparison between the classical results obtained from a Possio [10] solver and numerical results obtained from the present scheme for a .5 Mach number. The grid dimensions for the results shown in Figure 8 were 15 units in the streamwise direction and 10 units in the normal direction. The spacing in the  $x$  and  $y$  directions was uniform with  $\Delta x = \Delta y$ . There were 151 points in the  $x$  direction and 101 points in the  $y$  direction.

In Figure 9 we make the same comparison as in Figure 8, except in this case the grid dimensions have been extended to 22.5 by 15. The number of grid points in each coordinate direction is the same as in Figure 8, so that the spacing is slightly more coarse on this grid. The results shown in Figure 10 were run on a still larger grid, with dimensions of 30 by 22.5, and 201 points in the  $x$  direction and 151 points in the  $y$  direction.

By comparing the results shown in Figures 8-10, it is seen that the agreement for the high frequencies is very good in all three cases. However, the agreement for reduced frequencies below about .3 is only fair. Note that the amount of error in the low frequency cases diminishes as the grid dimensions become larger. This suggests that a larger grid is required to adequately model the long wavelength disturbances.



In Figures 11-13 we present results for the case of an .8 Mach number. The grids used for these figures correspond to the grids used in Figures 8-10, respectively. The agreement for the .8 Mach number case is analogous to the agreement for the .5 Mach number case, and again we see that the accuracy for lower frequencies is considerably improved on the larger grids.

## 6.2 Comparison of the Present Scheme with the Frequency Domain Approach of Scott and Atassi

As mentioned in the introduction, Scott and Atassi [5] have reported a general frequency domain approach that was developed to solve periodic vortical flows around isolated airfoils. Their scheme has wide applicability to many different flow configurations, and was recently extended to handle the difficult problem of vortical flows around lifting airfoils in subsonic flows [11]. Our purpose in the present section is to compare the present time domain calculation procedure with their frequency domain approach as specialized to the case of flat plate airfoils.

There are two main disadvantages to the frequency domain procedure reported in [5]. Perhaps the biggest disadvantage is that it requires the solution of a large matrix equation by a direct solver. Because the sparse solver described in [5] has been optimized for storage efficiency, it is not particularly computationally efficient. Large solution times can be required, on the order of 200 seconds per frequency on a Cray X-MP, as the reduced frequency becomes larger and the number of grid points increases. The other main disadvantage of the frequency domain scheme described in [5] is that there is a loss of accuracy as the Mach number and reduced frequency become large. Due to this loss of accuracy and the longer solution times for the higher frequencies, the frequency domain numerical scheme described by Scott and Atassi is not suitable for the very high frequencies, i.e., for reduced frequencies above about 5.0.

There are, however, a number of advantages to their frequency domain approach. It is very accurate for a large range of frequencies, including the very lowest frequencies. Their scheme is valid for both incompressible and compressible flows, and can handle three-dimensional gusts. In addition, the unsteady grid and far field boundary condition described in [5] are ideally suited for acoustic wave propagation, so that the acoustics can be accurately and readily determined from the solution. The scheme is also general and can be applied to airfoils with arbitrary geometry.

Among the advantages of the time domain scheme presented in this paper are that the scheme is very accurate at the higher frequencies, and it can handle nonperiodic vortical disturbances as easily as periodic ones. In addition, the present numerical approach is both storage and computationally efficient. For the special case of a flat plate airfoil in a transverse gust, the scheme is very efficient, as the airfoil response to a continuous spectrum of frequencies can be calculated simultaneously. Solution times for the entire response function range from 86 to 343 seconds on a Cray X-MP,

versus 600 to 1500 seconds for the same calculation using the frequency domain approach.

The improved computational efficiency of the present time domain approach is due entirely to the fact that solutions for a complete range of frequencies can be obtained simultaneously in a single calculation by Fourier decomposing the results of the airfoil response to a pulse disturbance. This approach carries with it the disadvantage that all of the harmonics have to be captured by a single grid. Since the long wavelength disturbances cannot be accurately resolved on a grid which is suitable for the high frequency cases, there is an inherent loss of accuracy for the low frequency disturbances. If, as in the frequency domain approach in [5], a separate calculation were performed on a separate grid for each reduced frequency, then the time domain scheme is actually less computationally efficient than the frequency domain scheme.

In addition to the loss of accuracy for low reduced frequencies, the other main disadvantage of the present time domain approach is that it is not valid for low Mach number flows.

### 6.3 Conclusion

In conclusion, the time domain numerical scheme which has been developed in the present paper is both storage and computationally efficient, and can accurately calculate the high frequency response of a flat plate airfoil in compressible flow to a transverse gust. Improvements are still needed in the accuracy of the calculation at the lower frequencies, and in the selection of a more suitable computational grid.

In as much as nonlinear equations such as the Euler equations are usually solved using time marching procedures, it is hoped that the present time domain solution procedure for the linearized, unsteady Euler equations will illuminate some of the issues involved in time domain calculations for unsteady flows.

## REFERENCES

1. Atassi, H.M.: Unsteady Vortical Disturbances Around Bodies. Proceedings of the Tenth U.S. National Congress of Applied Mechanics, J.P. Lamb, ed., ASME, 1986, pp 475-484.
2. Goldstein, M.E.: Unsteady Vortical and Entropic Distortions of Potential Flows Round Arbitrary Obstacles. J. Fluid Mech., vol. 89, pt. 3, Dec. 13, 1978, pp. 433-468.
3. Atassi, H.M.; and Grzedzinski, J.: Unsteady Disturbances of Streaming Motions Around Bodies. Accepted for publication in J. Fluid Mech., 1989.
4. Atassi, H.M.; and Scott, J.R.: Analysis of Nonuniform Subsonic Flows About a Row of Moving Blades. Proceedings of the Fourth International Symposium on Unsteady Aerodynamics and Aeroelasticity of Turbomachines and Propellers, H.E. Gallus and S. Servaty, eds., Institute fur Strahlantriebe und Turbomaschine, University of Aachen, Federal Republic of Germany, 1988, pp. 39-67.
5. Scott, J.R.; and Atassi, H.M.: Numerical Solution of Periodic Vortical Flows about a Thin Airfoil. AIAA Paper 89-1691, June 1989 (NASA TM-101998).
6. Bayliss, A.; and Turkel, E.: Far Field Boundary Conditions for Compressible Flows. J. Comput. Phys., vol 48, no. 2, Nov. 1982, pp. 182-199.
7. Friedlander, F.G.: On the Radiation Field of Pulse Solution of the Wave Equation. Proc. Roy. Soc. London Ser. A., vol. 269, no. 1336, Aug. 21, 1962, pp. 53-65.
8. Hariharan, S.I.; and Hagstrom, T.: Far Field Expansions for Anisotropic Wave Equations. NASA TM 102112, 1989.
9. Sears, W.R.: Some Aspects of Non-stationary Airfoil Theory and Its Practical Application. J. Aeronaut. Sci., vol 8, no. 3, Jan. 1941, pp. 104-108.
10. Possio, C.: L'Azione Aerodinamica sul Profilo Oscillante in un Fluido Compressibile a Velocita Ipsonara (Aerodynamic Action on Oscillating Profile in Compressible Fluid at Subsonic Velocities). L'Aerotechnica, vol. 18, no. 4, Apr. 1938, pp. 441-458.
11. Scott, J.R.: Compressible Flows with Periodic Vortical Disturbances Around Lifting Airfoils. Ph. D. dissertation in preparation, University of Notre Dame, Notre Dame, IN, 1989.

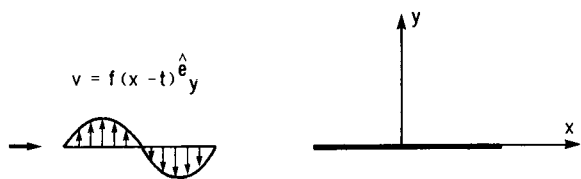


FIGURE 1. - FLAT PLATE AIRFOIL IN A TRANSVERSE GUST.

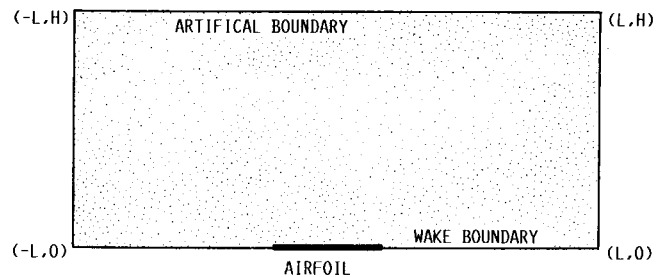


FIGURE 2. - COMPUTATIONAL DOMAIN.

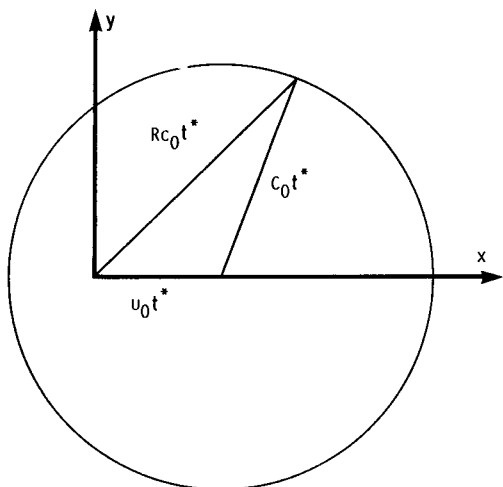


FIGURE 3. - WAVE FRONT WITH RESPECT TO A MOVING SOURCE.

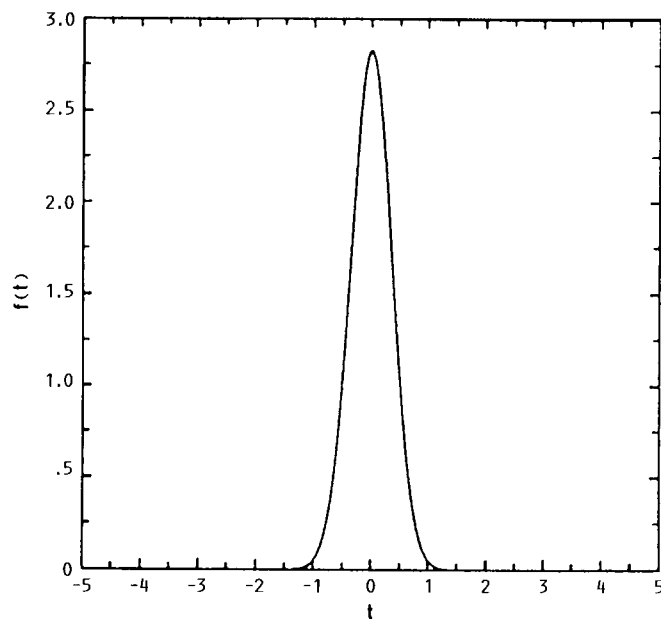


FIGURE 4. - GAUSSIAN PULSE.

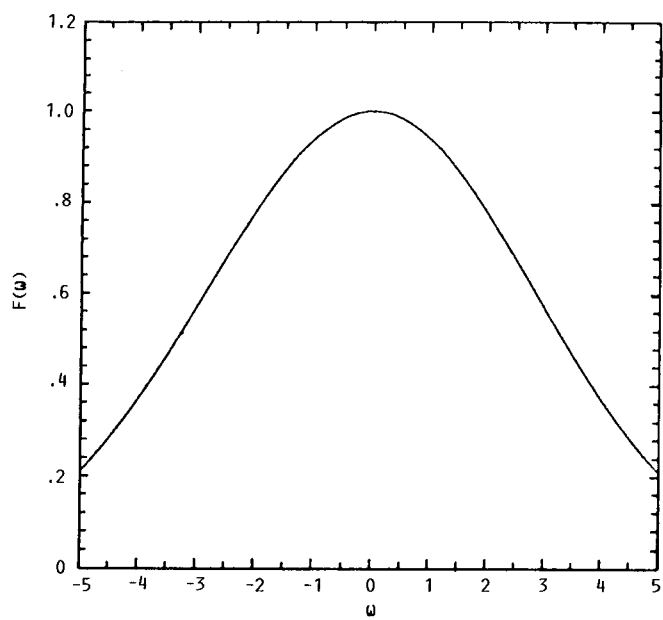


FIGURE 5. - FOURIER TRANSFORM OF  $f(t)$ .

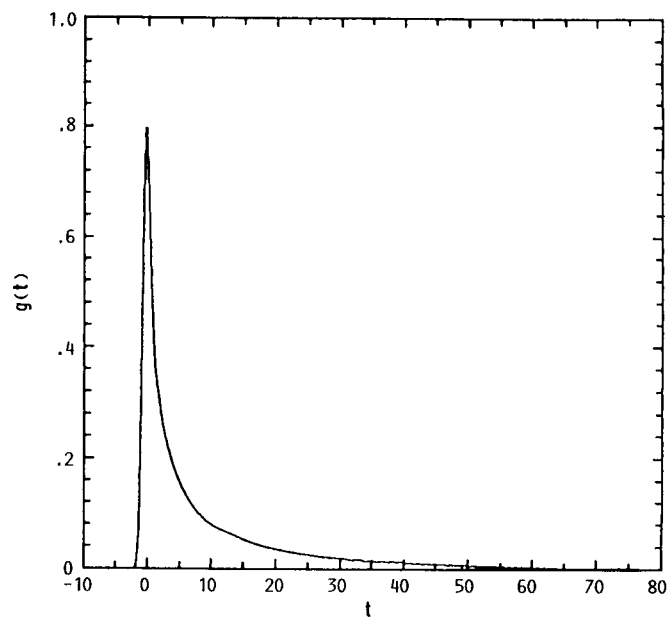


FIGURE 6. - RESPONSE FUNCTION.

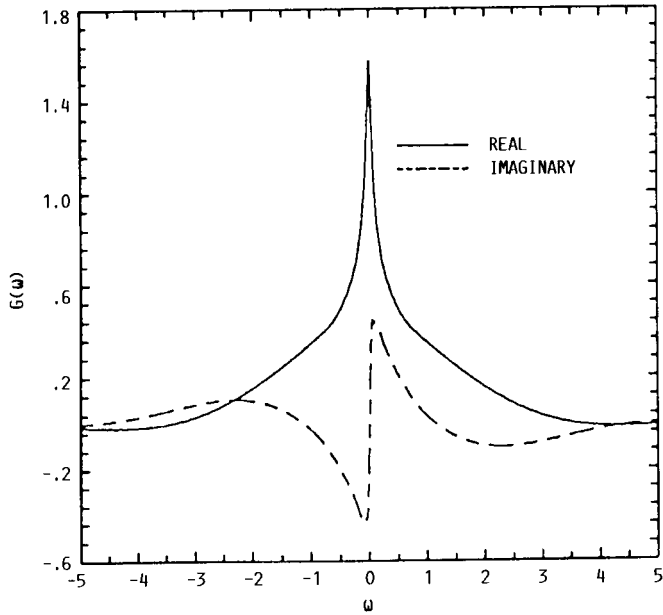


FIGURE 7. - FOURIER TRANSFORM OF  $g(t)$ .

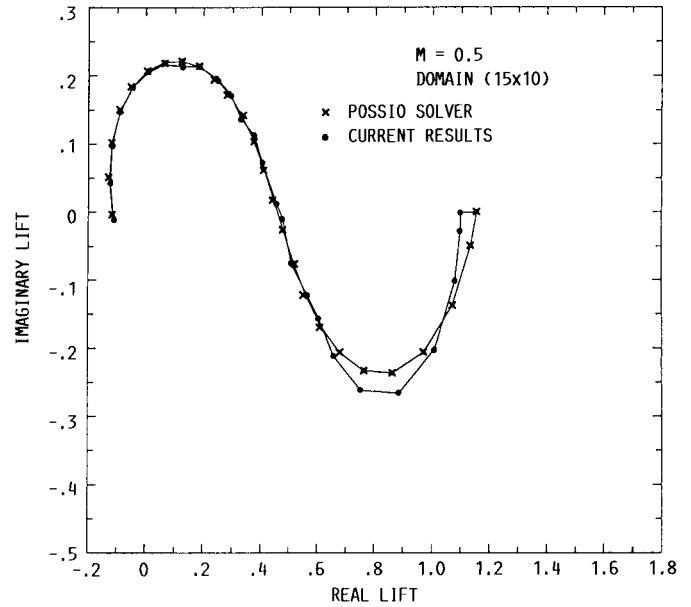


FIGURE 8. - COMPARISON BETWEEN THE NUMERICALLY COMPUTED UNSTEADY RESPONSE FUNCTION AND ANALYTICAL RESULTS FROM A POSSIO SOLVER.

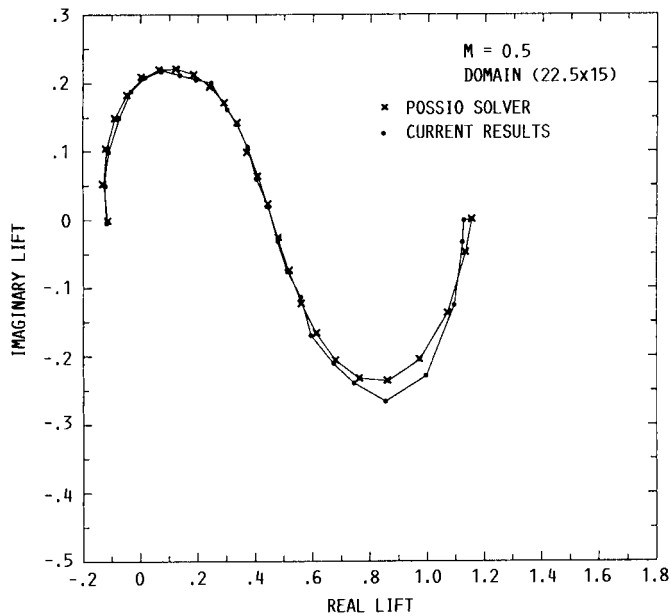


FIGURE 9. - COMPARISON BETWEEN THE NUMERICALLY COMPUTED UNSTEADY RESPONSE FUNCTION AND ANALYTICAL RESULTS FROM A POSSIO SOLVER.

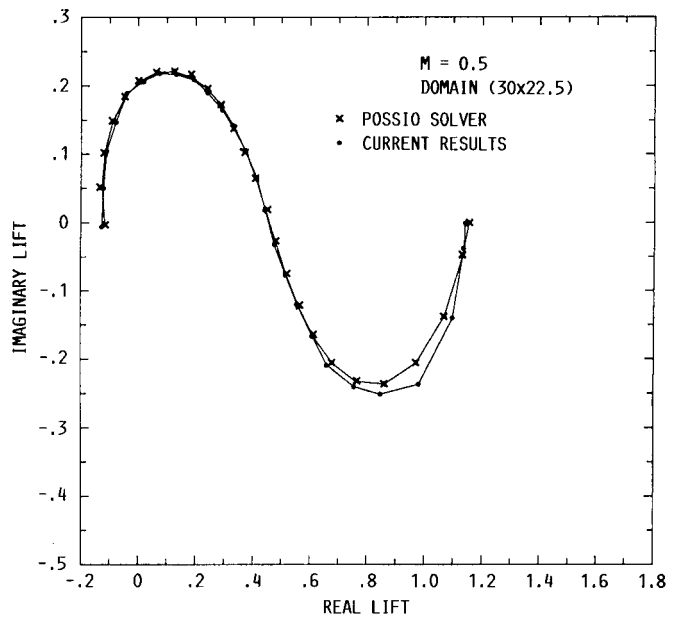


FIGURE 10. - COMPARISON BETWEEN THE NUMERICALLY COMPUTED UNSTEADY RESPONSE FUNCTION AND ANALYTICAL RESULTS FROM A POSSIO SOLVER.

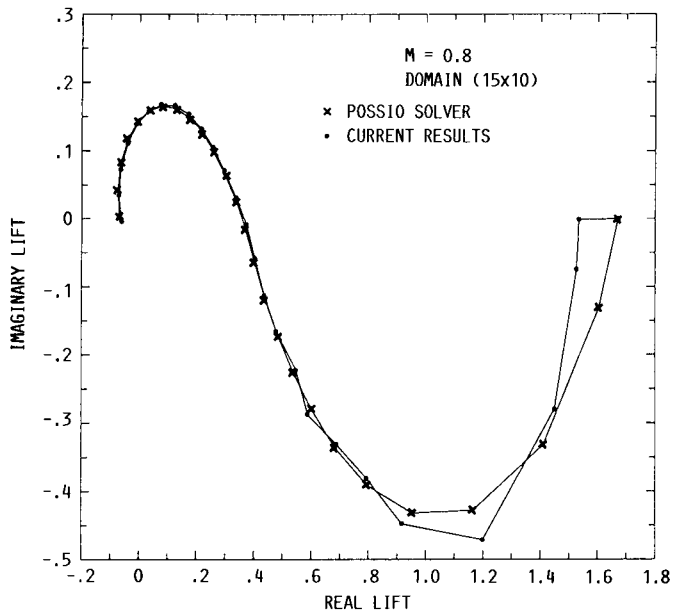


FIGURE 11. - COMPARISON BETWEEN THE NUMERICALLY COMPUTED UNSTEADY RESPONSE FUNCTION AND ANALYTICAL RESULTS FROM A POSSIO SOLVER.

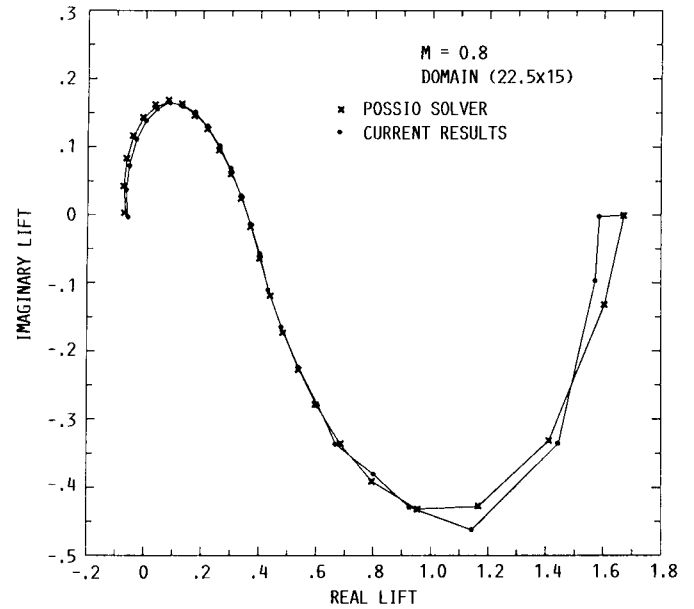


FIGURE 12. - COMPARISON BETWEEN THE NUMERICALLY COMPUTED UNSTEADY RESPONSE FUNCTION AND ANALYTICAL RESULTS FROM A POSSIO SOLVER.

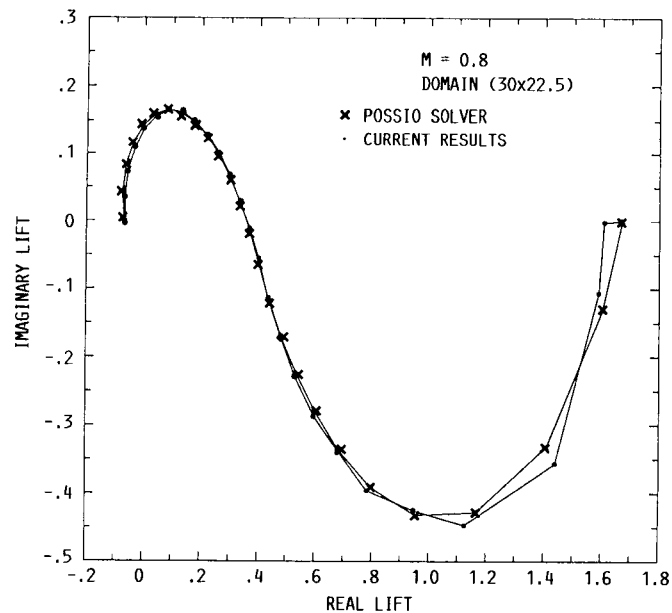


FIGURE 13. - COMPARISON BETWEEN THE NUMERICALLY COMPUTED UNSTEADY RESPONSE FUNCTION AND ANALYTICAL RESULTS FROM A POSSIO SOLVER.

# Report Documentation Page

1. Report No. NASA TM-102318 ICOMP-89-19		2. Government Accession No.		3. Recipient's Catalog No.	
4. Title and Subtitle Time Domain Numerical Calculations of Unsteady Vortical Flows About a Flat Plate Airfoil				5. Report Date September 1989	
				6. Performing Organization Code	
7. Author(s) S.I. Hariharan, Yu Ping, and J.R. Scott				8. Performing Organization Report No. E-5014	
				10. Work Unit No. 505-62-21	
9. Performing Organization Name and Address National Aeronautics and Space Administration Lewis Research Center Cleveland, Ohio 44135-3191				11. Contract or Grant No.	
				13. Type of Report and Period Covered Technical Memorandum	
12. Sponsoring Agency Name and Address National Aeronautics and Space Administration Washington, D.C. 20546-0001				14. Sponsoring Agency Code	
15. Supplementary Notes S.I. Hariharan, Department of Mathematical Sciences, University of Akron, Akron, Ohio 44325 and Institute for Computational Mechanics in Propulsion, NASA Lewis Research Center (work funded under Space Act Agreement C99066G). Space Act Monitor: Louis A. Povinelli. Yu Ping, Department of Mathematical Sciences, University of Akron. J.R. Scott, NASA Lewis Research Center.					
16. Abstract  A time domain numerical scheme is developed to solve for the unsteady flow about a flat plate airfoil due to imposed upstream, small amplitude, transverse velocity perturbations. The governing equation for the resulting unsteady potential is a homogeneous, constant coefficient, convective wave equation. Accurate solution of the problem requires the development of approximate boundary conditions which correctly model the physics of the unsteady flow in the far field. A uniformly valid far field boundary condition is developed, and numerical results are presented using this condition. The stability of the scheme is discussed, and the stability restriction for the scheme is established as a function of the Mach number. Finally, comparisons are made with the frequency domain calculation by Scott and Atassi, and the relative strengths and weaknesses of each approach are assessed.					
17. Key Words (Suggested by Author(s)) Vortical; Unsteady; Gust; Airfoil; Time domain				18. Distribution Statement Unclassified - Unlimited Subject Category 34	
19. Security Classif. (of this report) Unclassified		20. Security Classif. (of this page) Unclassified		21. No of pages 22	
				22. Price* A03	