NASA MEMO 5-12-59L

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BASIC PRESSURE MEASUREMENTS AT TRANSONIC SPEEDS ON A THIN 45° SWEPTBACK HIGHLY TAPERED WING WITH SYSTEMATIC SPANWISE TWIST VARIATIONS

WING WITH CUBIC SPANWISE TWIST VARIATION

By John P. Mugler, Jr.

Langley Research Center
Langley Field, Va.

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SUMMARY

Pressure distributions obtained in the Langley 8-foot transonic pressure tunnel on a thin, highly tapered, twisted, 45° sweptback wing in combination with a body are presented. The wing has a cubic spanwise twist variation from 0° at 10 percent of the semispan to 6° at the tip. The tip is at a lower angle of attack than the root. Tests were made at stagnation pressures of 1.0 and 0.5 atmosphere, at Mach numbers from 0.800 to 1.200, and at angles of attack from -4° to 20°.

INTRODUCTION

A research program has been conducted at the Langley Research Center to determine the loads due to wing twist at transonic and supersonic speeds. As part of this program, tests have been made in the Langley 8-foot transonic pressure tunnel on four wings; an untwisted wing to serve as a reference, and wings with linear, quadratic, and cubic variations of twist across the span. References 1, 2, and 3 present the basic pressure measurements on the untwisted, linearly twisted, and quadratically twisted wings, respectively. The present paper presents the basic pressure measurements on the wing with a cubic variation of twist across the span. These data are being presented without analysis.

SYMBOLS

b	wing span
b'/2	unsupported semispan (distance from outer face of wing mounting block to tip)

c	airfoil section chord, measured parallel to plane of symmetry
$\bar{c}$	wing mean aerodynamic chord
c_m	wing section pitching-moment coefficient about 0.25c, $\int_0^1 \left(C_{p,L} - C_{p,U}\right)\left(0.25 - \frac{x}{c}\right)d\left(\frac{x}{c}\right)$
c_n	wing section normal-force coefficient, $\int_0^1 \left(C_{p,L} - C_{p,U}\right)d\left(\frac{x}{c}\right)$
C_p	pressure coefficient
$C_{p,sonic}$	pressure coefficient corresponding to local Mach number of 1.0
D	diameter
l	body length
M	Mach number
q	free-stream dynamic pressure
R	Reynolds number based on $\bar{c}$
x	distance measured from leading edge of wing or from nose of body (positive rearward)
y	spanwise distance measured from body center line
y'	spanwise distance measured from outer face of wing mounting block
$\frac{\partial \Delta\alpha}{\partial n}$	wing-twist influence coefficient due to normal load at 0.25c
$\frac{\partial \Delta\alpha}{\partial m}$	wing-twist influence coefficient due to moment about 0.25c
α	angle of attack of wing-body center line
$\Delta\alpha$	Angle of attack of wing station - Angle of attack of wing-body center line

Subscripts:

L	lower surface
U	upper surface

APPARATUS

Tunnel

The investigation was conducted in the Langley 8-foot transonic pressure tunnel. The test section of this facility is rectangular in cross section. The upper and lower walls are slotted longitudinally to allow continuous operation through the transonic speed range with negligible effects of choking and blockage. During this investigation, the tunnel was operated at stagnation pressures of approximately 1.0 and 0.5 atmosphere. The dewpoint of the tunnel air was controlled and kept constant at approximately 0° F. The stagnation temperature of the tunnel air was automatically controlled and was kept constant and uniform across the tunnel at 123° F. Control of both dewpoint and stagnation temperature in this manner minimized humidity effects. Details of the test section have been presented in reference 4.

Models

The wing tested has the same plan form, thickness, and camber distribution as the untwisted wing described in reference 1. However, the wing of the present investigation had twist built into each wing panel from 10 percent of the semispan to the tip. The sections were twisted about the leading edge in planes parallel to the model plane of symmetry with the trailing edges up; therefore, the tips are at a lower angle of attack than the wing-body center line. The twist varied cubicly from 0° at the 10-percent-semispan station to 6° at the tip. The wing was constructed of steel and was tested as a midwing configuration. The wing was tested in combination with the basic body of reference 1. Details of the wing-body combination are presented in figure 1 and the wing twist characteristics are presented in table I.

TESTS

The wing-body combination was tested at Mach numbers from 0.800 to 1.200 at tunnel stagnation pressures of 1.0 and 0.5 atmosphere. At a

stagnation pressure of 0.5 atmosphere, the angles of attack ranged from -4° to 20°. For Mach numbers of 0.800, 0.900, and 0.940 at a stagnation pressure of 1.0 atmosphere, the angles of attack ranged from -4° to 20°; for other test Mach numbers at a stagnation pressure of 1.0 atmosphere, the angles of attack ranged from -4° to 12°.

Transition strips were fixed on the model during all the tests. The strips were about 0.10 inch wide and were formed by sprinkling No. 120 carborundum grains on a plastic adhesive. The strips extended from the wing-body juncture to the wing tip at 10 percent of the local chord on the upper and lower wing surfaces and formed a ring around the body at 10 percent of the body length.

The Reynolds number based on the wing mean aerodynamic chord varied over the Mach number range from about 2.6×10^6 to 2.9×10^6 during tests at 1.0 atmosphere and from about 1.3×10^6 to 1.5×10^6 during tests at 0.5 atmosphere. (See fig. 2.)

MEASUREMENTS AND ACCURACY

Measurements of the local static pressures on the model were made by using flush-mounted orifices distributed over the upper and lower wing surfaces and along longitudinal body rows. Figure 3 shows the location of the six stations on the wing and five rows on the body where the orifices were located. Pressure coefficients determined from these measurements are estimated to be accurate within ± 0.006.

The angle of attack of the model was measured by the use of a strain-gage attitude transmitter mounted in the nose of the model and is estimated to be accurate within $\pm 0.1^\circ$. Calibrations of the test section of the Langley 8-foot transonic pressure tunnel indicate that local deviations from the average free-stream Mach number are of the order of ± 0.005 at subsonic speeds. With increases in Mach number, these deviations increased but did not exceed ± 0.10 in the region of the wing at $M = 1.2$. Several representative Mach number distributions at the center of the test section have been presented in reference 4. The average stream Mach number was held to within ± 0.003 of the nominal values shown in the figures.

The stagnation pressures of 1,058 and 2,116 pounds per square foot have been designated 0.5 and 1.0 atmosphere, respectively, throughout this paper. The stagnation pressure was generally held to within ± 10 pounds per square foot during tests at 0.5 atmosphere and ± 20 pounds per square foot during tests at 1.0 atmosphere.

Influence coefficients were obtained for this wing from a static calibration and are presented in table II. Wing-twist angles, computed by using the experimental wing section data in conjunction with the influence coefficients of table II, are estimated to be accurate to within about $\pm 0.25^{\circ}$.

RESULTS

The pressure coefficients for the wing in the presence of the body are presented in tables III and IV for stagnation pressures of 0.5 and 1.0 atmosphere, respectively. Pressure coefficients for the body in the presence of the wing are presented in tables V and VI for stagnation pressures of 0.5 and 1.0 atmosphere, respectively. The values of the free-stream dynamic pressure shown in the tables is the average value over the angle-of-attack range. The pressure coefficients have been plotted to show the pressure-coefficient distributions over the surfaces and are presented in figure 4 for the wing and in figure 5 for the body. The distributions over the wing (fig. 4) have been numerically integrated for section normal-force and section pitching-moment coefficients about 0.25c and the results are presented in table VII. The section data were used in conjunction with the influence coefficients of table II to calculate the change in angle at several wing stations and these values are also presented in table VII.

In figures 4 and 5, data have been presented for both stagnation pressures on the same figure. Fixing transition during the tests tended to minimize the effects of Reynolds number on the pressure coefficients. This fact is evident from figures 4 and 5 which show that in all cases changing the stagnation pressure from 0.5 to 1.0 atmosphere had no significant effects on the pressure coefficients over the body or over the inboard wing stations. Aeroelastic effects caused the wing to twist over the outboard regions. The results in table VII show that the outboard wing sections are generally operating at a lesser angle of attack at a stagnation pressure of 1.0 atmosphere than at 0.5 atmosphere because of the differences in dynamic pressure. Therefore, the differences in the pressure distributions over the outboard wing sections at the two different stagnation pressures in figure 4 should be attributed to the differences in local angle of attack and not to Reynolds number effects.

Langley Research Center,
National Aeronautics and Space Administration,
Langley Field, Va., February 18, 1959.

REFERENCES

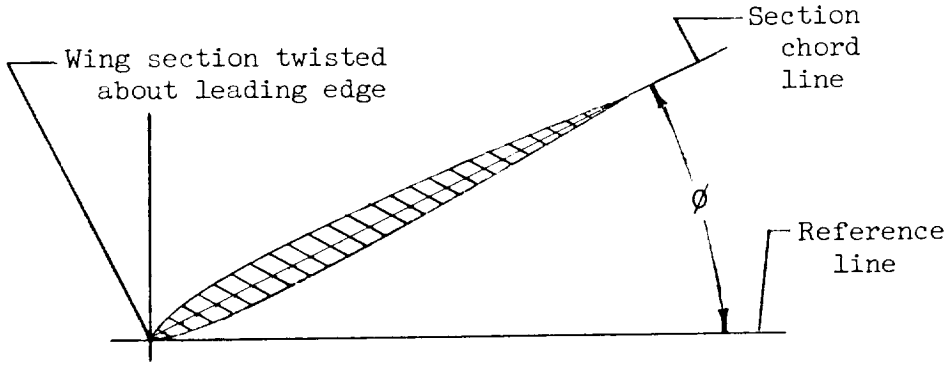
1. Mugler, John P., Jr.: Basic Pressure Measurements at Transonic Speeds on a Thin 45° Sweptback Highly Tapered Wing With Systematic Spanwise Twist Variations - Untwisted Wing. NASA MEMO 10-20-58L, 1958.

2. Mugler, John P., Jr.: Basic Pressure Measurements at Transonic Speeds on a Thin 45° Sweptback Highly Tapered Wing With Systematic Spanwise Twist Variations - Wing With Linear Spanwise Twist Variation. NASA MEMO 12-28-58L, 1959.

3. Mugler, John P., Jr.: Basic Pressure Measurements at Transonic Speeds on a Thin 45° Sweptback Highly Tapered Wing With Systematic Spanwise Twist Variations - Wing With Quadratic Spanwise Twist Variation. NASA MEMO 2-24-59L, 1959.

4. Mugler, John P., Jr.: Transonic Wind-Tunnel Investigation of the Aerodynamic Loading Characteristics of a 60° Delta Wing in the Presence of a Body With and Without Indentation. NACA RM L55G11, 1955.

TABLE I.- WING TWIST CHARACTERISTICS



Typical Section

$\frac{y}{b/2}$	ø, deg
0	0
.10	0
.12	.000
.25	.028
.40	.222
.60	1.029
.80	2.823
.95	5.054
1.00	6.000

TABLE II.- WING DEFLECTION CHARACTERISTICS

Twist measurement station, $\frac{y}{b/2}$	Rate of change in twist angle due to a load at section quarter chord, $\frac{\partial \Delta\alpha}{\partial n}$, deg/lb, at -				
	$\frac{y'}{b'/2} = 0.185$	$\frac{y'}{b'/2} = 0.348$	$\frac{y'}{b'/2} = 0.565$	$\frac{y'}{b'/2} = 0.795$	$\frac{y'}{b'/2} = 0.948$
0.25	0	-0.0003	-0.0006	-0.0008	-0.0013
.40	0	-.0002	-.0016	-.0026	-.0040
.60	0	-.0001	-.0016	-.0099	-.0169
.80	0	-.0001	-.0010	-.0119	-.0457
.95	0	-.0001	-.0008	-.0109	-.0547

Twist measurement station, $\frac{y}{b/2}$	Rate of change in twist angle due to a pitching moment about section quarter chord, $\frac{\partial \Delta\alpha}{\partial m}$, deg/in-lb, at -				
	$\frac{y'}{b'/2} = 0.185$	$\frac{y'}{b'/2} = 0.348$	$\frac{y'}{b'/2} = 0.565$	$\frac{y'}{b'/2} = 0.795$	$\frac{y'}{b'/2} = 0.948$
0.25	0.0001	0.0001	0.0001	0.0001	0
.40	.0002	.0003	.0002	.0003	0
.60	.0002	.0006	.0023	.0022	.0026
.80	.0002	.0006	.0036	.0124	.0131
.95	.0002	.0006	.0037	.0167	.0832

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.9 ATMOSPHERE FOR WING IN PRESENCE OF BODY

(a) 12-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.900; q = 310 lb/sq ft										
Upper surface	.000	.130	.348	.497	.523	.500	.246	-.101	-.384	-.643	.000
	.027	.337	.232	.115	-.016	-.172	-.542	-.864	-1.691	-1.440	.027
	.079	.172	.079	-.015	-.108	-.219	-.450	-.688	-.892	-1.332	.079
	.149	.093	.011	-.069	-.145	-.231	-.386	-.598	-.734	-1.120	.149
	.243	.038	-.030	-.102	-.164	-.239	-.390	-.549	-.491	-.844	.243
	.351	-.020	-.085	-.150	-.207	-.278	-.432	-.572	-.659	-.737	.351
	.447	-.063	-.122	-.181	-.233	-.300	-.449	-.525	-.589	-.653	.447
	.546	-.067	-.120	-.174	-.221	-.279	-.407	-.425	-.594	-.606	.546
	.643	-.082	-.129	-.177	-.213	-.266	-.365	-.436	-.570	-.608	.643
	.740	-.061	-.102	-.140	-.169	-.205	-.272	-.361	-.492	-.550	.740
	.820	-.062	-.093	-.122	-.141	-.168	-.212	-.292	-.443	-.554	.820
	.903	-.036	-.055	-.072	-.083	-.099	-.123	-.185	-.332	-.511	.903
Lower surface	.025	-.492	-.310	-.126	.028	.156	.385	.574	.726	.843	.025
	.068	-.263	-.164	-.059	.037	.122	.297	.459	.605	.725	.068
	.148	-.241	-.150	-.068	.011	.077	.225	.363	.490	.591	.148
	.251	-.232	-.158	-.082	-.015	.047	.178	.297	.407	.494	.251
	.350	-.223	-.160	-.093	-.029	.026	.142	.250	.351	.432	.350
	.449	-.229	-.172	-.111	-.053	-.003	.105	.202	.294	.371	.449
	.549	-.214	-.167	-.108	-.059	-.010	.087	.172	.255	.322	.549
	.647	-.186	-.148	-.100	-.056	-.017	.065	.135	.206	.266	.647
	.741										.741
	.825	-.098	-.075	-.046	-.014	.011	.067	.104	.142	.177	.825
	.882	-.064	-.048	-.025	.001	.021	.068	.091	.117	.139	.882
	M = 0.900; q = 354 lb/sq ft										
Upper surface	.000	.273	.423	.534	.551	.543	.387	.123	-.103	-.325	.000
	.027	.345	.245	.143	.014	-.115	-.421	-.794	-1.328	-1.305	.027
	.079	.178	.089	.003	-.095	-.183	-.386	-.574	-.745	-1.247	.079
	.149	.095	.015	-.060	-.140	-.212	-.352	-.521	-.616	-.851	.149
	.243	.039	-.029	-.097	-.167	-.225	-.332	-.481	-.589	-.633	.243
	.351	-.030	-.095	-.160	-.230	-.285	-.386	-.519	-.500	-.658	.351
	.447	-.082	-.147	-.217	-.284	-.340	-.442	-.558	-.500	-.663	.447
	.546	-.090	-.149	-.214	-.284	-.348	-.454	-.557	-.544	-.633	.546
	.643	-.113	-.169	-.235	-.321	-.392	-.516	-.560	-.619	-.646	.643
	.740	-.090	-.137	-.191	-.272	-.379	-.505	-.504	-.613	-.604	.740
	.820	-.088	-.125	-.163	-.212	-.307	-.505	-.451	-.592	-.601	.820
	.903	-.057	-.075	-.094	-.113	-.131	-.263	-.294	-.504	-.573	.903
Lower surface	.025	-.435	-.266	-.104	.027	.154	.393	.593	.748	.871	.025
	.068	-.226	-.144	-.048	.035	.125	.308	.478	.624	.746	.068
	.148	-.232	-.143	-.070	.001	.077	.236	.385	.509	.614	.148
	.251	-.240	-.162	-.092	-.030	.044	.187	.315	.428	.522	.251
	.350	-.245	-.176	-.109	-.056	.016	.148	.264	.371	.461	.350
	.449	-.283	-.210	-.138	-.085	-.020	.105	.213	.312	.402	.449
	.549	-.304	-.222	-.149	-.093	-.031	.083	.179	.272	.357	.549
	.647	-.293	-.202	-.138	-.091	-.037	.061	.139	.221	.301	.647
	.741										.741
	.825	-.133	-.099	-.064	-.037	.001	.061	.105	.157	.222	.825
	.882	-.085	-.063	-.037	-.017	.014	.060	.088	.130	.188	.882

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 367 lb/sq ft										
Upper surface	.000	.331	.459	.554	.576	.570	.438	.194	-.017	-.235	.000
	.027	.359	.259	.159	.045	-.079	-.361	-.724	-1.232	-1.345	.027
	.079	.187	.098	.014	-.068	-.153	-.338	-.524	-.696	-1.337	.079
	.149	.102	.021	-.050	-.115	-.185	-.311	-.458	-.560	-.682	.149
	.243	.045	-.024	-.090	-.144	-.198	-.299	-.440	-.529	-.602	.243
	.351	-.028	-.097	-.160	-.206	-.259	-.357	-.476	-.560	-.514	.351
	.447	-.090	-.164	-.220	-.268	-.315	-.406	-.517	-.596	-.622	.447
	.546	-.100	-.168	-.233	-.284	-.333	-.428	-.524	-.562	-.652	.546
	.643	-.136	-.218	-.283	-.332	-.384	-.479	-.579	-.476	-.681	.643
	.740	-.115	-.191	-.276	-.331	-.383	-.473	-.570	-.541	-.663	.740
	.820	-.114	-.169	-.281	-.350	-.409	-.502	-.564	-.561	-.667	.820
	.903	-.075	-.093	-.164	-.287	-.377	-.498	-.440	-.543	-.643	.903
Lower surface	.025	-.394	-.242	-.087	.048	.162	.394	.599	.762	.892	.025
	.068	-.208	-.125	-.037	.050	.132	.309	.487	.636	.768	.068
	.148	-.208	-.136	-.067	.010	.082	.236	.392	.524	.637	.148
	.251	-.226	-.152	-.091	-.023	.045	.184	.321	.444	.548	.251
	.350	-.234	-.175	-.115	-.049	.012	.143	.269	.389	.486	.350
	.449	-.269	-.211	-.158	-.092	-.029	.098	.219	.330	.428	.449
	.549	-.301	-.254	-.187	-.114	-.049	.074	.186	.289	.384	.549
	.647	-.324	-.278	-.206	-.119	-.060	.045	.143	.241	.329	.647
	.741										.741
	.825	-.320	-.187	-.081	-.050	-.023	.039	.110	.180	.253	.825
	.882	-.252	-.087	-.044	-.029	-.012	.033	.092	.156	.222	.882
	M = 0.980; q = 382 lb/sq ft										
Upper surface	.000	.384	.502	.585	.603	.602	.489	.264	.060	-.146	.000
	.027	.381	.292	.195	.082	-.033	-.305	-.640	-1.095	-1.222	.027
	.079	.211	.133	.048	-.030	-.113	-.287	-.442	-.589	-1.115	.079
	.149	.125	.056	-.019	-.083	-.146	-.264	-.379	-.484	-.569	.149
	.243	.067	.011	-.057	-.108	-.165	-.266	-.378	-.460	-.557	.243
	.351	-.010	-.063	-.120	-.171	-.221	-.311	-.410	-.502	-.599	.351
	.447	-.081	-.127	-.187	-.231	-.277	-.365	-.462	-.534	-.633	.447
	.546	-.091	-.144	-.205	-.251	-.297	-.389	-.471	-.559	-.653	.546
	.643	-.151	-.196	-.259	-.305	-.350	-.436	-.521	-.608	-.699	.643
	.740	-.151	-.198	-.260	-.304	-.353	-.438	-.527	-.607	-.654	.740
	.820	-.175	-.219	-.287	-.334	-.379	-.458	-.548	-.619	-.570	.820
	.903	-.175	-.229	-.289	-.336	-.385	-.466	-.551	-.560	-.403	.903
Lower surface	.025	-.339	-.184	-.051	.070	.184	.414	.620	.788	.921	.025
	.068	-.167	-.078	-.005	.074	.155	.330	.509	.665	.797	.068
	.148	-.175	-.102	-.040	.034	.103	.258	.416	.555	.668	.148
	.251	-.192	-.120	-.063	.000	.063	.206	.347	.476	.579	.251
	.350	-.207	-.136	-.087	-.031	.027	.162	.297	.420	.521	.350
	.449	-.245	-.180	-.131	-.080	-.022	.114	.244	.363	.464	.449
	.549	-.281	-.215	-.171	-.113	-.054	.086	.210	.322	.422	.549
	.647	-.305	-.252	-.204	-.149	-.085	.053	.167	.273	.367	.647
	.741										.741
	.825	-.316	-.252	-.206	-.143	-.056	.047	.136	.217	.298	.825
	.882	-.303	-.244	-.195	-.130	-.051	.040	.119	.196	.268	.882

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF

0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.000	.446	.556	.630	.641	.635	.547	.336	.137	-.065	.000
	.027	.417	.331	.237	.126	.015	-.226	-.543	-.996	-1.140	.027
	.079	.252	.213	.095	.017	-.061	-.212	-.352	-.512	-1.071	.079
	.149	.169	.099	.026	-.036	-.097	-.193	-.302	-.405	-.482	.149
	.243	.116	.057	-.013	-.069	-.116	-.197	-.300	-.380	-.477	.243
	.351	.043	-.015	-.069	-.123	-.168	-.237	-.333	-.425	-.520	.351
	.447	-.025	-.080	-.133	-.178	-.218	-.289	-.386	-.458	-.554	.447
	.546	-.034	-.095	-.153	-.200	-.238	-.314	-.398	-.485	-.578	.546
	.643	-.087	-.145	-.205	-.250	-.286	-.360	-.445	-.534	-.621	.643
	.740	-.093	-.148	-.203	-.250	-.287	-.365	-.452	-.540	-.611	.740
	.820	-.118	-.167	-.229	-.276	-.313	-.385	-.469	-.549	-.619	.820
	.903	-.126	-.171	-.232	-.283	-.320	-.395	-.478	-.552	-.613	.903
Lower surface	.025	-.267	-.129	.004	.120	.222	.459	.658	.833	.958	.025
	.068	-.111	-.029	.044	.118	.192	.379	.549	.715	.834	.068
	.148	-.126	-.063	.001	.069	.142	.305	.458	.603	.707	.148
	.251	-.133	-.079	-.024	.038	.105	.252	.392	.525	.620	.251
	.350	-.148	-.091	-.044	.011	.073	.211	.343	.470	.565	.350
	.449	-.181	-.136	-.081	-.031	.027	.163	.290	.415	.508	.449
	.549	-.214	-.166	-.115	-.065	-.001	.134	.257	.378	.465	.549
	.647	-.242	-.202	-.149	-.100	-.039	.099	.215	.328	.414	.647
	.741										.741
	.825	-.242	-.199	-.151	-.094	-.017	.097	.186	.275	.346	.825
	.882	-.234	-.189	-.141	-.085	-.009	.093	.172	.254	.319	.882
	M = 1.125; q = 425 lb/sq ft										
Upper surface	.000	.495	.565	.613	.625	.616	.557	.439	.387	.120	.000
	.027	.379	.296	.194	.093	-.009	-.225	-.455	-.772	-.930	.027
	.079	.224	.146	.065	-.007	-.077	-.234	-.338	-.413	-.891	.079
	.149	.167	.097	.028	-.039	-.093	-.190	-.284	-.330	-.383	.149
	.243	.133	.068	.008	-.050	-.099	-.185	-.253	-.290	-.347	.243
	.351	.065	.012	-.050	-.099	-.145	-.220	-.279	-.333	-.394	.351
	.447	.007	-.041	-.093	-.138	-.180	-.251	-.312	-.355	-.424	.447
	.546	-.009	-.062	-.113	-.160	-.200	-.269	-.342	-.383	-.451	.546
	.643	-.060	-.105	-.155	-.198	-.236	-.313	-.371	-.422	-.486	.643
	.740	-.052	-.103	-.157	-.203	-.240	-.313	-.370	-.425	-.478	.740
	.820	-.076	-.127	-.175	-.218	-.254	-.318	-.381	-.430	-.481	.820
	.903	-.088	-.135	-.189	-.229	-.262	-.331	-.394	-.436	-.472	.903
Lower surface	.025	-.228	-.118	-.026	.079	.167	.374	.605	.840	1.021	.025
	.068	-.127	-.058	.009	.090	.156	.317	.519	.718	.883	.068
	.148	-.110	-.058	-.007	.057	.122	.265	.442	.610	.754	.148
	.251	-.099	-.067	-.016	.041	.094	.232	.387	.542	.676	.251
	.350	-.100	-.071	-.023	.024	.082	.206	.345	.496	.627	.350
	.449	-.139	-.099	-.051	-.002	.052	.162	.304	.449	.579	.449
	.549	-.173	-.131	-.086	-.034	.015	.136	.276	.420	.545	.549
	.647	-.195	-.157	-.114	-.069	-.018	.096	.241	.381	.499	.647
	.741										.741
	.825	-.191	-.149	-.105	-.059	-.006	.118	.235	.349	.445	.825
	.882	-.182	-.136	-.098	-.049	.004	.125	.227	.336	.422	.882

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 443 lb/sq ft										
Upper surface	.000	.530	.591	.633	.642	.628	.579	.462	.370	.471	.000
	.027	.376	.290	.199	.103	.006	-.191	-.399	-.665	-.776	.027
	.079	.216	.150	.080	.006	-.058	-.213	-.320	-.363	-.728	.079
	.149	.139	.089	.029	-.023	-.077	-.165	-.265	-.320	-.410	.149
	.243	.145	.087	.022	-.035	-.080	-.162	-.222	-.270	-.314	.243
	.351	.086	.034	-.018	-.060	-.111	-.189	-.248	-.301	-.352	.351
	.447	.037	-.005	-.056	-.101	-.146	-.212	-.273	-.322	-.382	.447
	.546	.014	-.029	-.079	-.123	-.163	-.234	-.285	-.349	-.408	.546
	.643	-.028	-.074	-.122	-.163	-.200	-.268	-.328	-.389	-.447	.643
	.740	-.040	-.071	-.113	-.155	-.192	-.260	-.324	-.393	-.449	.740
	.820	-.058	-.096	-.135	-.173	-.208	-.279	-.340	-.399	-.453	.820
	.903	-.079	-.117	-.156	-.194	-.225	-.292	-.351	-.409	-.461	.903
Lower surface	.025	-.185	-.112	-.019	.085	.172	.359	.543	.753	.966	.025
	.068	-.096	-.044	.012	.087	.159	.300	.462	.667	.836	.068
	.148	-.088	-.053	-.000	.057	.113	.236	.406	.593	.730	.148
	.251	-.083	-.051	-.009	.043	.104	.233	.373	.531	.660	.251
	.350	-.085	-.050	-.007	.044	.096	.213	.343	.485	.608	.350
	.449	-.118	-.076	-.030	.021	.071	.181	.301	.438	.560	.449
	.549	-.146	-.111	-.067	-.018	.035	.157	.267	.409	.526	.549
	.647	-.168	-.137	-.094	-.047	.008	.105	.212	.370	.481	.647
	.741										.741
	.825	-.158	-.135	-.097	-.052	-.007	.096	.232	.352	.439	.825
	.882	-.155	-.121	-.091	-.055	-.009	.101	.237	.346	.422	.882

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.000	-.012	.164	.310	.391	.238	-.591	-1.344	-1.265	-1.051	.000
	.024	.334	.237	.099	-.098	-.373	-1.331	-1.460	-1.191	-1.067	.024
	.075	.157	.062	-.059	-.187	-.348	-.661	-1.258	-1.214	-1.003	.075
	.152	.078	-.010	-.107	-.205	-.324	-.564	-1.135	-1.165	-.932	.152
	.252	-.010	-.092	-.179	-.259	-.354	-.558	-.874	-1.159	-.941	.252
	.353	-.048	-.120	-.197	-.263	-.345	-.525	-.547	-1.042	-.909	.353
	.448	-.089	-.157	-.222	-.280	-.353	-.503	-.467	-.918	-.867	.448
	.552	-.080	-.136	-.188	-.235	-.291	-.394	-.460	-.792	-.819	.552
	.652	-.074	-.119	-.160	-.197	-.233	-.305	-.414	-.690	-.777	.652
	.759	-.047	-.085	-.110	-.133	-.163	-.204	-.322	-.595	-.744	.759
	.858	-.013	-.037	-.052	-.066	-.081	-.111	-.222	-.503	-.706	.858
	.935	.027	.015	.005	.004	-.007	-.029	-.112	-.397	-.664	.935
Lower surface	.027	-.652	-.442	-.225	.001	.171	.389	.499	.572	.609	.027
	.077	-.603	-.304	-.110	.009	.115	.289	.416	.526	.601	.077
	.154	-.490	-.212	-.086	-.001	.078	.229	.346	.450	.535	.154
	.252	-.246	-.182	-.097	-.020	.044	.172	.280	.377	.460	.252
	.347	-.222	-.175	-.102	-.034	.019	.133	.232	.329	.410	.347
	.447	-.196	-.159	-.099	-.040	.008	.112	.197	.275	.347	.447
	.559	-.169	-.136	-.086	-.036	.003	.093	.161	.228	.288	.559
	.658	-.125	-.098	-.057	-.016	.017	.090	.144	.195	.241	.658
	.756	-.082	-.061	-.029	.001	.027	.085	.120	.151	.184	.756
	.862	-.028	-.018	.004	.026	.043	.080	.089	.089	.094	.862
	.908	-.002	.005	.021	.038	.050	.074	.070	.044	.032	.908
	M = 0.900; q = 354 lb/sq ft										
Upper surface	.000	.072	.209	.323	.404	.309	-.224	-.904	-1.229	-1.019	.000
	.024	.326	.236	.111	-.069	-.299	-1.023	-1.392	-1.197	-.965	.024
	.075	.148	.055	-.055	-.181	-.320	-.634	-1.282	-1.114	-.947	.075
	.152	.067	-.021	-.111	-.211	-.311	-.495	-.805	-.990	-.875	.152
	.252	-.028	-.113	-.199	-.290	-.369	-.524	-.689	-.993	-.886	.252
	.353	-.074	-.149	-.232	-.315	-.389	-.525	-.618	-.936	-.873	.353
	.448	-.122	-.196	-.290	-.384	-.454	-.587	-.631	-.869	-.848	.448
	.552	-.112	-.175	-.247	-.355	-.447	-.579	-.608	-.794	-.810	.552
	.652	-.104	-.156	-.213	-.292	-.441	-.583	-.575	-.729	-.773	.652
	.759	-.071	-.107	-.142	-.174	-.255	-.533	-.509	-.672	-.743	.759
	.858	-.028	-.050	-.068	-.084	-.071	-.168	-.353	-.620	-.718	.858
	.935	.022	.010	.005	-.004	.007	-.046	-.223	-.547	-.690	.935
Lower surface	.027	-.596	-.443	-.241	-.031	.148	.386	.516	.595	.647	.027
	.077	-.470	-.253	-.105	-.008	.109	.296	.430	.539	.619	.077
	.154	-.412	-.219	-.101	-.027	.068	.228	.358	.462	.561	.154
	.252	-.343	-.216	-.121	-.051	.027	.169	.287	.391	.485	.252
	.347	-.310	-.210	-.130	-.068	.008	.137	.241	.342	.426	.347
	.447	-.319	-.217	-.136	-.077	-.012	.104	.200	.290	.377	.447
	.559	-.285	-.184	-.120	-.072	-.016	.085	.165	.242	.322	.559
	.658	-.181	-.130	-.083	-.044	.002	.082	.143	.210	.281	.658
	.756	-.111	-.081	-.047	-.018	.020	.079	.117	.168	.229	.756
	.862	-.040	-.025	-.005	.016	.039	.071	.079	.100	.149	.862
	.908	-.010	.003	.017	.030	.049	.063	.050	.054	.091	.908

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 367 lb/sq ft										
Upper surface	.000	.116	.227	.330	.420	.347	-.104	-.741	-1.155	-1.219	.000
	.024	.327	.237	.117	-.039	-.255	-.877	-1.276	-1.213	-1.124	.024
	.075	.151	.060	-.045	-.155	-.282	-.607	-1.160	-1.185	-1.056	.075
	.152	.066	-.018	-.107	-.190	-.282	-.448	-.679	-1.071	-.948	.152
	.252	-.031	-.119	-.200	-.268	-.347	-.477	-.621	-1.075	-.966	.252
	.353	-.083	-.162	-.239	-.300	-.365	-.486	-.617	-.952	-.966	.353
	.448	-.147	-.248	-.315	-.373	-.430	-.547	-.632	-.849	-.934	.448
	.552	-.140	-.224	-.315	-.374	-.439	-.546	-.619	-.753	-.883	.552
	.652	-.132	-.208	-.324	-.387	-.450	-.557	-.608	-.710	-.839	.652
	.759	-.091	-.138	-.284	-.369	-.431	-.547	-.571	-.672	-.795	.759
	.858	-.038	-.053	-.062	-.261	-.392	-.516	-.486	-.634	-.755	.858
	.935	.020	.015	.021	-.014	-.130	-.258	-.290	-.514	-.693	.935
Lower surface	.027	-.586	-.464	-.249	-.027	.142	.380	.523	.610	.671	.027
	.077	-.470	-.255	-.115	-.006	.101	.283	.433	.554	.650	.077
	.154	-.361	-.221	-.105	-.023	.062	.223	.364	.477	.585	.154
	.252	-.329	-.213	-.139	-.055	.017	.163	.292	.407	.511	.252
	.347	-.314	-.239	-.163	-.085	-.011	.116	.242	.362	.459	.347
	.447	-.319	-.258	-.184	-.102	-.034	.091	.206	.307	.405	.447
	.559	-.343	-.283	-.187	-.095	-.037	.067	.168	.263	.352	.559
	.658	-.333	-.260	-.116	-.061	-.020	.062	.149	.231	.313	.658
	.756	-.309	-.124	-.056	-.029	-.006	.052	.124	.191	.263	.756
	.862	-.109	-.014	-.002	.009	.010	.031	.079	.128	.188	.862
	.908	-.022	.015	.023	.029	.017	.012	.047	.084	.135	.908
	M = 0.980; q = 382 lb/sq ft										
Upper surface	.000	.166	.268	.357	.444	.382	-.013	-.614	-1.025	-1.177	.000
	.024	.346	.265	.150	.001	-.199	-.787	-1.183	-1.146	-1.221	.024
	.075	.166	.090	-.012	-.119	-.241	-.589	-1.112	-1.104	-1.179	.075
	.152	.081	.014	-.072	-.153	-.242	-.400	-.615	-.960	-1.117	.152
	.252	-.020	-.087	-.164	-.233	-.305	-.431	-.554	-1.026	-1.181	.252
	.353	-.074	-.132	-.207	-.263	-.328	-.445	-.554	-.986	-1.117	.353
	.448	-.164	-.211	-.281	-.334	-.393	-.496	-.597	-.849	-.985	.448
	.552	-.157	-.217	-.293	-.348	-.403	-.501	-.599	-.616	-.858	.552
	.652	-.174	-.238	-.312	-.366	-.419	-.513	-.611	-.591	-.810	.652
	.759	-.177	-.236	-.305	-.357	-.408	-.508	-.602	-.536	-.771	.759
	.858	-.167	-.222	-.291	-.342	-.394	-.493	-.575	-.574	-.737	.858
	.935	-.135	-.172	-.228	-.272	-.310	-.366	-.422	-.506	-.572	.935
Lower surface	.027	-.564	-.450	-.225	-.013	.152	.394	.545	.640	.704	.027
	.077	-.407	-.188	-.077	.020	.120	.308	.455	.579	.676	.077
	.154	-.332	-.172	-.081	-.003	.077	.240	.386	.506	.618	.154
	.252	-.303	-.178	-.110	-.042	.026	.179	.315	.436	.545	.252
	.347	-.279	-.190	-.136	-.071	-.007	.141	.268	.388	.490	.347
	.447	-.291	-.224	-.171	-.112	-.047	.103	.228	.338	.442	.447
	.559	-.318	-.254	-.206	-.143	-.067	.076	.191	.295	.392	.559
	.658	-.318	-.254	-.202	-.136	-.052	.070	.174	.266	.355	.658
	.756	-.310	-.244	-.193	-.129	-.046	.056	.147	.230	.308	.756
	.862	-.294	-.232	-.182	-.120	-.041	.033	.103	.172	.236	.862
	.908	-.277	-.214	-.170	-.111	-.047	.008	.068	.128	.184	.908

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.000	.227	.316	.396	.475	.427	.077	-.490	-.921	-1.107	.000
	.024	.386	.301	.188	.045	-.140	-.672	-1.032	-1.071	-1.137	.024
	.075	.212	.129	.032	-.073	-.181	-.495	-.948	-1.022	-1.091	.075
	.152	.131	.057	-.028	-.110	-.188	-.322	-.554	-.778	-1.024	.152
	.252	.035	-.038	-.115	-.184	-.247	-.349	-.463	-.783	-1.119	.252
	.353	-.016	-.080	-.150	-.213	-.269	-.366	-.472	-.697	-1.110	.353
	.448	-.100	-.156	-.221	-.279	-.329	-.416	-.513	-.726	-1.107	.448
	.552	-.100	-.164	-.230	-.292	-.336	-.421	-.514	-.697	-.802	.552
	.652	-.120	-.186	-.253	-.309	-.352	-.434	-.525	-.670	-.671	.652
	.759	-.121	-.179	-.246	-.301	-.344	-.432	-.524	-.644	-.656	.759
	.858	-.115	-.167	-.237	-.288	-.329	-.420	-.515	-.585	-.304	.858
	.935	-.092	-.126	-.192	-.242	-.270	-.322	-.431	-.311	-.231	.935
Lower surface	.027	-.506	-.442	-.180	.024	.189	.436	.581	.688	.743	.027
	.077	-.356	-.166	-.047	.050	.154	.346	.498	.629	.715	.077
	.154	-.265	-.132	-.040	.040	.122	.285	.429	.556	.654	.154
	.252	-.240	-.136	-.060	.005	.075	.224	.359	.488	.585	.252
	.347	-.228	-.155	-.088	-.026	.035	.180	.313	.442	.533	.347
	.447	-.222	-.170	-.114	-.059	.004	.147	.272	.392	.485	.447
	.559	-.248	-.202	-.149	-.093	-.021	.123	.238	.350	.435	.559
	.658	-.246	-.201	-.148	-.089	-.014	.117	.221	.322	.400	.658
	.756	-.237	-.192	-.139	-.080	-.008	.107	.197	.287	.357	.756
	.862	-.224	-.179	-.130	-.072	-.002	.086	.156	.231	.289	.862
	.908	-.218	-.169	-.125	-.070	-.007	.065	.123	.189	.241	.908
	M = 1.125; q = 425 lb/sq ft										
Upper surface	.000	.286	.353	.419	.480	.454	.220	-.262	-.674	-.880	.000
	.024	.389	.313	.196	.059	-.109	-.556	-.842	-.913	-.932	.024
	.075	.226	.149	.049	-.058	-.162	-.460	-.785	-.889	-.886	.075
	.152	.154	.083	.006	-.075	-.146	-.305	-.613	-.703	-.815	.152
	.252	.062	-.006	-.073	-.139	-.194	-.294	-.373	-.687	-.901	.252
	.353	.018	-.040	-.100	-.156	-.213	-.315	-.378	-.698	-.911	.353
	.448	-.056	-.111	-.169	-.219	-.267	-.353	-.429	-.668	-.928	.448
	.552	-.070	-.123	-.183	-.231	-.271	-.350	-.428	-.540	-.708	.552
	.652	-.088	-.135	-.191	-.239	-.286	-.371	-.441	-.468	-.544	.652
	.759	-.080	-.135	-.188	-.241	-.284	-.368	-.435	-.451	-.507	.759
	.858	-.074	-.130	-.183	-.231	-.274	-.356	-.419	-.412	-.091	.858
	.935	-.060	-.114	-.160	-.207	-.244	-.291	-.363	-.311	-.174	.935
Lower surface	.027	-.523	-.453	-.173	.010	.166	.404	.582	.712	.803	.027
	.077	-.271	-.116	-.047	.064	.160	.333	.497	.649	.774	.077
	.154	-.160	-.082	-.007	.067	.138	.280	.439	.586	.716	.154
	.252	-.160	-.086	-.018	.043	.100	.225	.373	.521	.655	.252
	.347	-.148	-.098	-.042	.019	.073	.196	.332	.483	.608	.347
	.447	-.167	-.121	-.071	-.019	.034	.155	.298	.442	.567	.447
	.559	-.187	-.150	-.102	-.050	.005	.128	.271	.409	.525	.559
	.658	-.187	-.154	-.106	-.053	.002	.131	.260	.391	.496	.658
	.756	-.184	-.150	-.105	-.050	.007	.129	.249	.366	.459	.756
	.862	-.178	-.142	-.097	-.046	.008	.126	.221	.322	.400	.862
	.908	-.175	-.139	-.095	-.043	.010	.116	.199	.284	.354	.908

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 443 lb/sq ft										
Upper surface	.000	.331	.390	.448	.505	.482	.305	-.067	-.482	-.710	.000
	.024	.386	.304	.210	.083	-.065	-.408	-.634	-.811	-.832	.024
	.075	.223	.143	.063	-.026	-.130	-.364	-.565	-.780	-.751	.075
	.152	.158	.102	.014	-.061	-.126	-.285	-.524	-.615	-.678	.152
	.252	.068	.017	-.042	-.104	-.158	-.276	-.310	-.543	-.743	.252
	.353	.032	-.010	-.068	-.127	-.186	-.280	-.339	-.450	-.737	.353
	.448	-.039	-.086	-.137	-.191	-.239	-.315	-.376	-.462	-.762	.448
	.552	-.055	-.101	-.152	-.200	-.242	-.314	-.376	-.468	-.707	.552
	.652	-.076	-.120	-.164	-.206	-.252	-.334	-.400	-.470	-.581	.652
	.759	-.075	-.121	-.166	-.212	-.253	-.327	-.389	-.470	-.500	.759
	.858	-.072	-.112	-.157	-.203	-.246	-.320	-.376	-.466	-.486	.858
	.935	-.063	-.098	-.145	-.187	-.229	-.295	-.332	-.433	-.447	.935
Lower surface	.027	-.466	-.413	-.188	.007	.153	.387	.560	.699	.796	.027
	.077	-.185	-.114	-.056	.036	.138	.310	.458	.630	.754	.077
	.154	-.137	-.084	-.021	.051	.129	.279	.419	.568	.697	.154
	.252	-.128	-.081	-.025	.044	.112	.230	.357	.504	.634	.252
	.347	-.117	-.091	-.045	.012	.068	.180	.300	.464	.588	.347
	.447	-.159	-.111	-.061	-.006	.046	.167	.282	.429	.549	.447
	.559	-.177	-.139	-.091	-.037	.016	.125	.246	.400	.512	.559
	.658	-.183	-.147	-.100	-.047	.004	.123	.244	.388	.489	.658
	.756	-.164	-.126	-.097	-.049	.004	.114	.250	.372	.459	.756
	.862	-.151	-.114	-.070	-.021	.030	.132	.247	.342	.413	.862
	.908	-.143	-.117	-.076	-.028	.024	.124	.227	.314	.376	.908

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.000	.292	.314	.300	.101	-.887	-1.493	-1.258	-1.028	-.934	.000
	.029	.293	.195	.043	-.187	-.517	-1.362	-1.241	-.970	-.842	.029
	.077	.166	.071	-.067	-.227	-.419	-1.081	-1.156	-.982	-.835	.077
	.151	.082	-.003	-.109	-.227	-.377	-.904	-1.106	-.953	-.821	.151
	.249	.021	-.056	-.149	-.243	-.356	-.662	-1.090	-.900	-.800	.249
	.348	-.025	-.099	-.176	-.256	-.351	-.529	-1.047	-.868	-.781	.348
	.439	-.044	-.107	-.175	-.240	-.319	-.439	-.954	-.843	-.766	.439
	.549	-.034	-.088	-.144	-.189	-.257	-.333	-.793	-.807	-.750	.549
	.646	-.053	-.096	-.139	-.176	-.224	-.272	-.733	-.768	-.732	.646
	.743	-.044	-.077	-.109	-.131	-.166	-.196	-.604	-.725	-.712	.743
	.841	-.011	-.036	-.055	-.068	-.089	-.109	-.452	-.682	-.693	.841
	.914	.024	.007	-.005	-.008	-.018	-.039	-.317	-.638	-.670	.914
Lower surface	.026	-.673	-.540	-.223	.032	.209	.393	.477	.518	.539	.026
	.077	-.606	-.382	-.131	.007	.126	.293	.404	.480	.538	.077
	.149	-.616	-.291	-.076	.001	.091	.237	.342	.430	.491	.149
	.244	-.556	-.209	-.093	-.017	.054	.184	.274	.353	.424	.244
	.345	-.321	-.168	-.086	-.020	.040	.152	.232	.302	.365	.345
	.444	-.184	-.136	-.074	-.014	.035	.126	.193	.257	.310	.444
	.542	-.124	-.100	-.050	-.000	.038	.118	.169	.217	.262	.542
	.638	-.072	-.058	-.019	.020	.054	.118	.151	.184	.218	.638
	.731	-.031	-.022	.011	.041	.066	.114	.130	.141	.163	.731
	.828	.013	.019	.042	.066	.081	.113	.107	.082	.087	.828
	.877	.029	.032	.051	.072	.082	.107	.086	.034	.031	.877
	M = 0.900; q = 394 lb/sq ft										
Upper surface	.000	.310	.319	.295	.170	-.556	-1.417	-1.184	-.858	-.966	.000
	.029	.270	.174	.032	-.179	-.474	-1.288	-1.138	-.840	-.828	.029
	.077	.145	.046	-.082	-.247	-.438	-1.066	-1.107	-.862	-.831	.077
	.151	.062	-.028	-.140	-.276	-.406	-.688	-1.055	-.874	-.838	.151
	.249	-.003	-.087	-.187	-.310	-.420	-.634	-.998	-.850	-.827	.249
	.348	-.056	-.134	-.227	-.331	-.453	-.653	-.928	-.825	-.803	.348
	.439	-.075	-.143	-.227	-.362	-.463	-.657	-.870	-.810	-.785	.439
	.549	-.063	-.124	-.195	-.290	-.433	-.625	-.782	-.790	-.772	.549
	.646	-.077	-.124	-.175	-.228	-.405	-.639	-.743	-.762	-.758	.646
	.743	-.063	-.096	-.130	-.163	-.151	-.306	-.691	-.733	-.739	.743
	.841	-.024	-.045	-.064	-.081	-.074	-.115	-.630	-.706	-.719	.841
	.914	.017	.005	-.004	-.015	-.006	-.045	-.558	-.678	-.698	.914
Lower surface	.026	-.697	-.581	-.262	-.018	.169	.382	.486	.539	.568	.026
	.077	-.586	-.370	-.159	-.034	.097	.282	.404	.491	.560	.077
	.149	-.549	-.295	-.108	-.041	.065	.230	.345	.438	.512	.149
	.244	-.510	-.275	-.132	-.058	.032	.173	.277	.366	.449	.244
	.345	-.438	-.230	-.124	-.057	.020	.138	.230	.310	.391	.345
	.444	-.328	-.181	-.104	-.046	.015	.116	.192	.265	.341	.444
	.542	-.221	-.129	-.073	-.027	.024	.105	.164	.228	.299	.542
	.638	-.132	-.075	-.032	.004	.045	.106	.146	.196	.258	.638
	.731	-.070	-.027	.002	.030	.063	.106	.123	.156	.210	.731
	.828	-.010	.021	.041	.061	.086	.107	.091	.102	.143	.828
	.877	.011	.037	.053	.070	.091	.100	.061	.060	.094	.877

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 367 lb/sq ft										
Upper surface	.000	.323	.327	.292	.216	-.379	-1.253	-1.268	-1.011	-.855	.000
	.029	.263	.162	.031	-.146	-.415	-1.143	-1.257	-.972	-.846	.029
	.077	.133	.032	-.090	-.229	-.391	-1.040	-1.257	-.943	-.864	.077
	.151	.049	-.048	-.152	-.257	-.383	-.597	-1.127	-.903	-.898	.151
	.249	-.020	-.113	-.215	-.302	-.401	-.575	-1.050	-.866	-.910	.249
	.348	-.078	-.171	-.259	-.349	-.441	-.590	-1.006	-.822	-.889	.348
	.439	-.097	-.202	-.298	-.372	-.463	-.609	-.931	-.810	-.868	.439
	.549	-.090	-.167	-.288	-.379	-.442	-.570	-.774	-.826	-.854	.549
	.646	-.095	-.157	-.309	-.402	-.485	-.620	-.732	-.813	-.837	.646
	.743	-.073	-.102	-.144	-.393	-.477	-.619	-.666	-.786	-.814	.743
	.841	-.025	-.044	-.042	-.103	-.367	-.421	-.587	-.761	-.791	.841
	.914	.024	.011	.010	.015	-.070	-.195	-.508	-.732	-.766	.914
Lower surface	.026	-.710	-.588	-.311	-.035	.148	.371	.487	.554	.592	.026
	.077	-.577	-.390	-.185	-.046	.078	.269	.405	.507	.582	.077
	.149	-.531	-.302	-.135	-.049	.046	.212	.346	.454	.536	.149
	.244	-.477	-.305	-.185	-.077	.007	.158	.279	.383	.472	.244
	.345	-.447	-.301	-.191	-.081	-.007	.121	.231	.330	.419	.345
	.444	-.413	-.308	-.160	-.066	-.011	.096	.195	.283	.369	.444
	.542	-.383	-.267	-.093	-.042	.001	.084	.167	.248	.325	.542
	.638	-.323	-.102	-.039	-.006	.022	.081	.151	.218	.289	.638
	.731	-.169	-.014	.004	.027	.042	.076	.126	.181	.243	.731
	.828	-.018	.038	.047	.065	.067	.071	.100	.134	.182	.828
	.877	.018	.052	.060	.076	.074	.059	.074	.097	.137	.877
	M = 0.980; q = 382 lb/sq ft										
Upper surface	.000	.345	.354	.313	.255	-.228	-1.198	-1.268	-1.120	-1.170	.000
	.029	.269	.185	.058	-.097	-.354	-1.065	-1.245	-1.106	-1.150	.029
	.077	.140	.055	-.062	-.189	-.336	-.961	-1.230	-1.090	-1.148	.077
	.151	.051	-.027	-.121	-.221	-.337	-.575	-1.146	-1.085	-1.148	.151
	.249	-.024	-.098	-.188	-.270	-.360	-.518	-.905	-1.074	-1.139	.249
	.348	-.093	-.146	-.242	-.325	-.400	-.533	-.821	-1.064	-1.087	.348
	.439	-.138	-.196	-.275	-.348	-.428	-.564	-.778	-1.077	-.996	.439
	.549	-.138	-.199	-.275	-.352	-.412	-.525	-.697	-1.054	-.940	.549
	.646	-.180	-.245	-.323	-.392	-.454	-.577	-.686	-.966	-.928	.646
	.743	-.193	-.256	-.330	-.391	-.453	-.576	-.650	-.823	-.897	.743
	.841	-.184	-.255	-.329	-.388	-.446	-.563	-.621	-.795	-.861	.841
	.914	-.112	-.169	-.223	-.281	-.348	-.403	-.454	-.751	-.826	.914
Lower surface	.026	-.674	-.526	-.304	-.033	.143	.380	.506	.580	.623	.026
	.077	-.539	-.344	-.164	-.043	.077	.277	.422	.531	.615	.077
	.149	-.488	-.253	-.116	-.037	.042	.224	.366	.480	.569	.149
	.244	-.434	-.261	-.168	-.092	-.007	.166	.298	.411	.507	.244
	.345	-.413	-.263	-.189	-.115	-.038	.128	.253	.361	.457	.345
	.444	-.392	-.277	-.215	-.138	-.041	.100	.215	.318	.409	.444
	.542	-.372	-.275	-.209	-.130	-.038	.086	.190	.284	.369	.542
	.638	-.340	-.252	-.191	-.114	-.022	.080	.173	.257	.332	.638
	.731	-.311	-.232	-.171	-.094	-.008	.071	.148	.225	.288	.731
	.828	-.254	-.178	-.120	-.048	.009	.058	.119	.184	.231	.828
	.877	-.195	-.131	-.083	-.025	.010	.040	.092	.153	.187	.877

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF

0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.000	.394	.400	.359	.296	-.146	-1.001	-1.149	-1.079	-1.110	.000
	.029	.314	.224	.108	-.048	-.300	-.898	-1.099	-1.076	-1.088	.029
	.077	.187	.095	-.015	-.140	-.270	-.826	-1.079	-1.057	-1.090	.077
	.151	.102	.057	-.072	-.171	-.275	-.547	-1.007	-1.059	-1.109	.151
	.249	.033	-.049	-.129	-.215	-.295	-.429	-.774	-1.069	-1.136	.249
	.348	-.030	-.103	-.190	-.268	-.333	-.448	-.663	-1.068	-1.137	.348
	.439	-.079	-.145	-.219	-.295	-.362	-.483	-.616	-1.063	-1.142	.439
	.549	-.082	-.152	-.218	-.293	-.350	-.448	-.584	-1.069	-1.113	.549
	.646	-.127	-.190	-.269	-.333	-.390	-.495	-.607	-1.063	-1.058	.646
	.743	-.143	-.198	-.273	-.335	-.390	-.498	-.607	-1.019	-.912	.743
	.841	-.144	-.202	-.270	-.332	-.382	-.490	-.599	-.964	-.816	.841
	.914	-.101	-.148	-.211	-.271	-.325	-.373	-.492	-.694	-.817	.914
Lower surface	.026	-.609	-.475	-.254	.007	.185	.417	.544	.628	.659	.026
	.077	-.474	-.302	-.120	-.001	.123	.319	.461	.580	.652	.077
	.149	-.421	-.212	-.069	.006	.091	.264	.405	.531	.608	.149
	.244	-.361	-.214	-.113	-.043	.043	.209	.339	.462	.548	.244
	.345	-.335	-.210	-.130	-.063	.009	.170	.296	.413	.496	.345
	.444	-.317	-.226	-.157	-.089	-.003	.145	.259	.370	.451	.444
	.542	-.299	-.222	-.155	-.083	.002	.133	.235	.338	.414	.542
	.638	-.269	-.199	-.137	-.067	.014	.129	.219	.312	.378	.638
	.731	-.246	-.179	-.121	-.050	.030	.122	.196	.280	.338	.731
	.828	-.203	-.137	-.080	-.011	.049	.110	.168	.241	.286	.828
	.877	-.169	-.103	-.053	.004	.049	.094	.140	.214	.248	.877
	M = 1.125; q = 425 lb/sq ft										
Upper surface	.000	.420	.423	.390	.344	.036	-.838	-.997	-.888	-.919	.000
	.029	.322	.246	.140	.004	-.225	-.737	-.956	-.885	-.910	.029
	.077	.199	.123	.024	-.087	-.223	-.660	-.917	-.884	-.905	.077
	.151	.124	.054	-.040	-.128	-.227	-.566	-.873	-.890	-.922	.151
	.249	.061	-.006	-.086	-.167	-.253	-.370	-.826	-.906	-.949	.249
	.348	.000	-.074	-.144	-.214	-.282	-.385	-.744	-.894	-.950	.348
	.439	-.041	-.099	-.173	-.241	-.306	-.427	-.519	-.874	-.946	.439
	.549	-.046	-.099	-.158	-.225	-.297	-.378	-.428	-.870	-.937	.549
	.646	-.092	-.152	-.215	-.274	-.323	-.424	-.487	-.845	-.824	.646
	.743	-.102	-.160	-.222	-.281	-.332	-.433	-.489	-.781	-.691	.743
	.841	-.106	-.164	-.224	-.278	-.327	-.423	-.479	-.648	-.693	.841
	.914	-.091	-.138	-.200	-.251	-.298	-.365	-.417	-.549	-.612	.914
Lower surface	.026	-.547	-.409	-.228	.002	.183	.401	.554	.662	.732	.026
	.077	-.413	-.275	-.100	.016	.128	.310	.471	.612	.723	.077
	.149	-.310	-.164	-.044	.021	.104	.263	.421	.569	.683	.149
	.244	-.266	-.170	-.085	-.008	.058	.210	.362	.507	.628	.244
	.345	-.268	-.164	-.091	-.025	.034	.171	.325	.467	.584	.345
	.444	-.266	-.177	-.115	-.051	.010	.158	.295	.434	.545	.444
	.542	-.243	-.175	-.116	-.052	.010	.150	.278	.408	.510	.542
	.638	-.218	-.159	-.104	-.039	.025	.158	.271	.390	.482	.638
	.731	-.196	-.135	-.082	-.021	.039	.163	.258	.366	.446	.731
	.828	-.160	-.101	-.050	.006	.064	.169	.240	.334	.400	.828
	.877	-.140	-.085	-.034	.020	.071	.161	.218	.306	.363	.877

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 443 lb/sq ft										
Upper surface	.000	.467	.461	.429	.383	.137	-.577	-.749	-.840	-.825	.000
	.029	.347	.277	.188	.032	-.155	-.536	-.710	-.833	-.824	.029
	.077	.224	.153	.068	-.036	-.186	-.508	-.689	-.830	-.820	.077
	.151	.141	.077	-.002	-.084	-.180	-.469	-.668	-.808	-.820	.151
	.249	.081	.018	-.045	-.123	-.209	-.415	-.639	-.717	-.813	.249
	.348	.009	-.041	-.103	-.169	-.232	-.325	-.597	-.657	-.800	.348
	.439	-.022	-.075	-.139	-.207	-.274	-.361	-.563	-.634	-.800	.439
	.549	-.028	-.068	-.123	-.191	-.265	-.339	-.394	-.631	-.800	.549
	.646	-.086	-.132	-.187	-.238	-.288	-.369	-.418	-.639	-.801	.646
	.743	-.099	-.143	-.194	-.248	-.296	-.376	-.430	-.649	-.794	.743
	.841	-.107	-.145	-.192	-.244	-.291	-.373	-.434	-.652	-.803	.841
	.914	-.092	-.128	-.172	-.219	-.263	-.345	-.410	-.611	-.774	.914
Lower surface	.026	-.539	-.511	-.248	.005	.186	.414	.549	.661	.730	.026
	.077	-.416	-.274	-.103	.036	.142	.325	.460	.602	.707	.077
	.149	-.208	-.113	-.023	.041	.122	.276	.403	.557	.667	.149
	.244	-.173	-.127	-.056	.007	.077	.215	.342	.494	.608	.244
	.345	-.184	-.133	-.070	-.006	.051	.176	.298	.453	.564	.345
	.444	-.212	-.161	-.103	-.042	.017	.137	.273	.423	.527	.444
	.542	-.217	-.172	-.113	-.050	.014	.126	.272	.405	.498	.542
	.638	-.188	-.156	-.109	-.047	.011	.123	.274	.392	.476	.638
	.731	-.163	-.122	-.075	-.029	.019	.142	.269	.373	.447	.731
	.828	-.136	-.096	-.045	.010	.062	.183	.266	.350	.410	.828
	.877	-.124	-.086	-.037	.019	.071	.184	.252	.326	.377	.877

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF

0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.000	.077	.159	.289	.084	-.720	-1.009	-.758	-.700	-.684	.000
	.026	.354	.244	.047	-.330	-.823	-.998	-.749	-.703	-.694	.026
	.077	.225	.120	-.028	-.235	-.520	-.952	-.731	-.682	-.676	.077
	.148	.134	.035	-.092	-.239	-.425	-.933	-.739	-.679	-.682	.148
	.251	.065	-.019	-.123	-.233	-.368	-.936	-.723	-.665	-.672	.251
	.350	.019	-.056	-.144	-.235	-.334	-.909	-.698	-.650	-.663	.350
	.452										.452
	.553	-.020	-.076	-.138	-.200	-.257	-.774	-.642	-.621	-.644	.553
	.647	-.036	-.082	-.131	-.175	-.223	-.647	-.616	-.605	-.632	.647
	.752	-.040	-.078	-.116	-.141	-.170	-.483	-.582	-.586	-.620	.752
	.854	-.007	-.037	-.061	-.071	-.095	-.319	-.555	-.568	-.608	.854
	.907	.015	-.011	-.026	-.032	-.051	-.228	-.538	-.559	-.602	.907
	.925										.925
Lower surface	.040	-.598	-.465	-.184	.076	.235	.390	.446	.473	.493	.040
	.092	-.597	-.442	-.107	.052	.170	.313	.386	.439	.480	.092
	.153	-.596	-.414	-.110	.018	.116	.249	.324	.380	.432	.153
	.251	-.592	-.305	-.065	.025	.100	.210	.272	.322	.371	.251
	.343	-.582	-.209	-.058	.016	.081	.174	.226	.268	.317	.343
	.452	-.516	-.125	-.037	.020	.074	.148	.181	.216	.259	.452
	.546	-.379	-.066	-.014	.034	.076	.136	.149	.171	.208	.546
	.651	-.177	-.027	.016	.052	.085	.128	.115	.128	.154	.651
	.799	.056	.020	.046	.074	.088	.102	.038	.032	.050	.799
	.878	.095	.033	.050	.074	.077	.078	-.035	-.050	-.040	.878
	M = 0.900; q = 354 lb/sq ft										
Upper surface	.000	.062	.164	.287	.107	-.636	-1.220	-.805	-.713	-.714	.000
	.026	.334	.224	.030	-.366	-.967	-1.244	-.743	-.721	-.723	.026
	.077	.208	.098	-.055	-.299	-.631	-1.200	-.720	-.702	-.703	.077
	.148	.115	.013	-.120	-.325	-.491	-1.093	-.722	-.696	-.705	.148
	.251	.048	-.042	-.153	-.304	-.500	-.964	-.713	-.684	-.699	.251
	.350	.004	-.078	-.174	-.302	-.534	-.882	-.700	-.674	-.690	.350
	.452										.452
	.553	-.036	-.093	-.160	-.246	-.236	-.715	-.675	-.651	-.676	.553
	.647	-.052	-.103	-.155	-.218	-.206	-.624	-.660	-.637	-.668	.647
	.752	-.053	-.095	-.133	-.179	-.173	-.523	-.634	-.624	-.658	.752
	.854	-.015	-.048	-.066	-.084	-.089	-.406	-.605	-.612	-.654	.854
	.907	.012	-.016	-.028	-.038	-.043	-.339	-.585	-.605	-.649	.907
	.925										.925
Lower surface	.040	-.691	-.506	-.237	.031	.200	.366	.444	.482	.510	.040
	.092	-.651	-.426	-.119	.020	.147	.297	.381	.443	.494	.092
	.153	-.645	-.402	-.132	-.013	.095	.234	.319	.382	.444	.153
	.251	-.671	-.346	-.088	-.001	.084	.194	.266	.332	.392	.251
	.343	-.639	-.235	-.075	-.005	.070	.164	.221	.280	.340	.343
	.452	-.586	-.139	-.049	.004	.066	.139	.179	.228	.285	.452
	.546	-.454	-.080	-.023	.023	.072	.128	.148	.188	.241	.546
	.651	-.151	-.033	.011	.049	.084	.120	.116	.146	.191	.651
	.799	.117	.016	.045	.072	.091	.093	.043	.060	.094	.799
	.878	.113	.032	.052	.070	.083	.060	-.025	-.013	.016	.878

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 367 lb/sq ft										
Upper surface	.000	.063	.143	.256	.179	-.400	-1.152	-1.073	-.801	-.821	.000
	.026	.304	.188	.004	-.301	-.807	-1.313	-.774	-.812	-.856	.026
	.077	.178	.064	-.095	-.271	-.612	-1.212	-.765	-.775	-.799	.077
	.148	.090	-.022	-.179	-.326	-.462	-1.168	-.774	-.749	-.773	.148
	.251	.024	-.073	-.229	-.349	-.473	-1.098	-.765	-.740	-.770	.251
	.350	-.018	-.101	-.260	-.400	-.507	-.880	-.745	-.731	-.760	.350
	.452										.452
	.553	-.047	-.100	-.149	-.425	-.556	-.659	-.700	-.709	-.741	.553
	.647	-.059	-.109	-.153	-.414	-.546	-.595	-.686	-.698	-.730	.647
	.752	-.058	-.100	-.136	-.115	-.528	-.547	-.664	-.687	-.719	.752
	.854	-.018	-.048	-.067	-.013	-.118	-.281	-.647	-.673	-.715	.854
	.907	.006	-.015	-.025	.012	-.044	-.224	-.633	-.666	-.711	.907
	.925										.925
Lower surface	.040	-.725	-.617	-.347	-.010	.164	.342	.442	.491	.529	.040
	.092	-.670	-.517	-.189	-.012	.114	.269	.379	.456	.518	.092
	.153	-.657	-.474	-.194	-.045	.062	.203	.315	.398	.468	.153
	.251	-.645	-.371	-.113	-.023	.053	.172	.266	.345	.419	.251
	.343	-.632	-.275	-.093	-.021	.042	.137	.219	.297	.368	.343
	.452	-.569	-.156	-.056	-.002	.045	.113	.178	.249	.315	.452
	.546	-.428	-.090	-.024	.023	.055	.104	.152	.209	.273	.546
	.651	-.270	-.044	.014	.057	.073	.099	.123	.171	.225	.651
	.799	-.090	.009	.050	.084	.087	.078	.057	.091	.136	.799
	.878	-.017	.029	.057	.088	.084	.052	-.004	.025	.058	.878
	M = 0.980; q = 382 lb/sq ft										
Upper surface	.000	.068	.159	.256	.253	-.185	-1.098	-1.226	-.901	-.813	.000
	.026	.278	.176	.025	-.223	-.693	-1.242	-1.152	-.942	-.805	.026
	.077	.147	.047	-.073	-.226	-.491	-1.113	-1.086	-.859	-.787	.077
	.148	.050	-.049	-.163	-.288	-.431	-1.077	-1.036	-.836	-.810	.148
	.251	-.029	-.117	-.222	-.324	-.431	-.998	-1.031	-.834	-.828	.251
	.350	-.088	-.169	-.323	-.371	-.464	-.931	-.999	-.804	-.834	.350
	.452										.452
	.553	-.152	-.229	-.322	-.423	-.515	-.596	-.914	-.768	-.821	.553
	.647	-.187	-.258	-.350	-.424	-.517	-.641	-.862	-.753	-.806	.647
	.752	-.230	-.306	-.398	-.494	-.567	-.696	-.806	-.736	-.789	.752
	.854	-.089	-.250	-.366	-.466	-.554	-.687	-.753	-.740	-.785	.854
	.907	-.033	-.101	-.163	-.337	-.441	-.655	-.726	-.733	-.779	.907
	.925										.925
Lower surface	.040	-.665	-.555	-.373	-.083	.112	.334	.452	.520	.562	.040
	.092	-.589	-.451	-.214	-.066	.078	.269	.394	.485	.546	.092
	.153	-.575	-.422	-.231	-.108	.018	.200	.331	.426	.499	.153
	.251	-.567	-.371	-.217	-.111	.006	.162	.281	.378	.452	.251
	.343	-.544	-.345	-.233	-.130	-.012	.128	.238	.331	.403	.343
	.452	-.530	-.330	-.228	-.128	-.016	.099	.198	.284	.354	.452
	.546	-.493	-.305	-.205	-.109	.000	.084	.174	.248	.313	.546
	.651	-.419	-.267	-.164	-.066	.015	.073	.150	.212	.267	.651
	.799	-.267	-.136	-.043	.014	.016	.036	.094	.137	.180	.799
	.878	-.180	-.056	.008	.025	-.001	-.000	.040	.070	.109	.878

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.000	.125	.209	.303	.294	-.087	-.871	-1.116	-.994	-.922	.000
	.026	.318	.224	.076	-.167	-.612	-1.078	-1.180	-1.006	-.934	.026
	.077	.192	.099	-.016	-.169	-.415	-.955	-1.126	-.948	-.906	.077
	.148	.101	.004	-.107	-.229	-.363	-.925	-1.126	-.955	-.930	.148
	.251	.027	-.068	-.160	-.264	-.367	-.870	-1.102	-.951	-.916	.251
	.350	-.036	-.122	-.217	-.306	-.388	-.822	-1.077	-.921	-.879	.350
	.452										.452
	.553	-.101	-.175	-.262	-.361	-.444	-.508	-1.012	-.848	-.829	.553
	.647	-.133	-.202	-.283	-.362	-.442	-.543	-.966	-.820	-.807	.647
	.752	-.181	-.247	-.344	-.425	-.489	-.600	-.917	-.793	-.787	.752
	.854	-.161	-.236	-.327	-.412	-.486	-.599	-.880	-.778	-.786	.854
	.907	-.112	-.179	-.290	-.383	-.452	-.587	-.788	-.768	-.778	.907
	.925										.925
Lower surface	.040	-.586	-.477	-.312	-.028	.157	.371	.485	.569	.602	.040
	.092	-.519	-.417	-.167	-.027	.114	.303	.427	.533	.588	.092
	.153	-.504	-.380	-.177	-.062	.054	.238	.366	.477	.543	.153
	.251	-.480	-.302	-.155	-.061	.046	.206	.316	.428	.497	.251
	.343	-.460	-.289	-.176	-.084	.019	.169	.275	.383	.451	.343
	.452	-.432	-.280	-.174	-.083	.014	.146	.239	.337	.402	.452
	.546	-.399	-.251	-.152	-.065	.034	.133	.213	.304	.363	.546
	.651	-.361	-.220	-.118	-.034	.051	.123	.189	.269	.322	.651
	.799	-.275	-.141	-.050	.024	.054	.089	.133	.197	.236	.799
	.878	-.213	-.095	-.030	.018	.031	.050	.085	.136	.169	.878
	M = 1.125; q = 425 lb/sq ft										
Upper surface	.000	.193	.264	.343	.364	.038	-.702	-.950	-.804	-.779	.000
	.026	.353	.268	.136	-.081	-.476	-.920	-1.007	-.800	-.764	.026
	.077	.234	.147	.048	-.098	-.299	-.787	-.933	-.748	-.734	.077
	.148	.138	.052	-.040	-.154	-.291	-.762	-.940	-.743	-.752	.148
	.251	.062	-.015	-.098	-.190	-.298	-.719	-.915	-.745	-.729	.251
	.350	.003	-.072	-.155	-.230	-.319	-.679	-.895	-.725	-.699	.350
	.452										.452
	.553	-.060	-.131	-.216	-.300	-.363	-.423	-.864	-.670	-.644	.553
	.647	-.096	-.151	-.223	-.305	-.376	-.439	-.852	-.642	-.633	.647
	.752	-.138	-.213	-.290	-.355	-.413	-.509	-.822	-.619	-.619	.752
	.854	-.137	-.206	-.281	-.355	-.412	-.509	-.763	-.602	-.600	.854
	.907	-.125	-.189	-.270	-.343	-.388	-.502	-.687	-.592	-.593	.907
	.925										.925
Lower surface	.040	-.491	-.403	-.273	.007	.163	.368	.516	.624	.688	.040
	.092	-.402	-.340	-.108	.026	.137	.314	.461	.590	.674	.092
	.153	-.394	-.307	-.118	-.008	.080	.245	.403	.538	.632	.153
	.251	-.381	-.236	-.109	-.025	.057	.217	.359	.498	.591	.251
	.343	-.335	-.216	-.129	-.047	.027	.190	.326	.456	.549	.343
	.452	-.306	-.215	-.140	-.064	.010	.174	.296	.419	.505	.452
	.546	-.289	-.207	-.128	-.062	.020	.174	.278	.392	.472	.546
	.651	-.275	-.193	-.108	-.041	.043	.176	.261	.366	.434	.651
	.799	-.231	-.134	-.054	.010	.086	.158	.214	.304	.357	.799
	.878	-.195	-.110	-.040	.025	.071	.125	.172	.251	.296	.878

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 443 lb/sq ft										
Upper surface	.000	.204	.278	.342	.396	.127	-.433	-.657	-.836	-.804	.000
	.026	.351	.273	.163	-.027	-.350	-.683	-.800	-.859	-.799	.026
	.077	.231	.163	.071	-.059	-.244	-.593	-.726	-.808	-.754	.077
	.148	.133	.068	-.017	-.124	-.253	-.584	-.726	-.853	-.808	.148
	.251	.066	.000	-.068	-.161	-.258	-.578	-.722	-.858	-.815	.251
	.350	.011	-.053	-.120	-.195	-.277	-.559	-.708	-.852	-.800	.350
	.452										.452
	.553	-.044	-.108	-.176	-.244	-.309	-.529	-.688	-.832	-.754	.553
	.647	-.070	-.115	-.179	-.257	-.325	-.479	-.659	-.808	-.735	.647
	.752	-.134	-.180	-.234	-.295	-.358	-.396	-.622	-.686	-.687	.752
	.854	-.125	-.181	-.250	-.312	-.362	-.434	-.662	-.603	-.692	.854
	.907	-.115	-.175	-.238	-.309	-.339	-.432	-.651	-.570	-.689	.907
	.925										.925
Lower surface	.040	-.502	-.511	-.332	-.022	.158	.353	.503	.619	.680	.040
	.092	-.374	-.395	-.197	-.016	.117	.293	.440	.581	.662	.092
	.153	-.346	-.310	-.148	-.027	.076	.223	.380	.529	.617	.153
	.251	-.336	-.205	-.100	-.016	.061	.208	.363	.492	.580	.251
	.343	-.267	-.191	-.117	-.044	.023	.182	.327	.453	.539	.343
	.452	-.238	-.182	-.112	-.043	.023	.177	.307	.419	.499	.452
	.546	-.227	-.160	-.088	-.027	.036	.189	.307	.396	.470	.546
	.651	-.214	-.149	-.077	-.013	.055	.198	.295	.371	.435	.651
	.799	-.169	-.105	-.033	.035	.113	.205	.256	.315	.365	.799
	.878	-.144	-.088	-.021	.047	.113	.179	.217	.269	.312	.878

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.034	.362	.275	.142	-.135	-.768	-.564	-.488	-.505	-.566	.034
	.082	.271	.180	.052	-.154	-.529	-.548	-.475	-.497	-.560	.082
	.162	.188	.099	-.008	-.171	-.393	-.529	-.465	-.488	-.553	.162
	.252	.119	.045	-.059	-.183	-.340	-.503	-.454	-.481	-.549	.252
	.347	.069	.008	-.079	-.184	-.308	-.475	-.443	-.472	-.544	.347
	.443	.030	-.024	-.101	-.192	-.288	-.443	-.427	-.459	-.535	.443
	.544	-.011	-.052	-.122	-.193	-.267	-.420	-.421	-.455	-.532	.544
	.589	-.012	-.048	-.112	-.174	-.237	-.406	-.415	-.453	-.530	.589
	.644	-.026	-.055	-.109	-.161	-.214	-.392	-.412	-.452	-.527	.644
	.738	-.043	-.051	-.092	-.130	-.163	-.377	-.404	-.442	-.520	.738
	.839	-.053	-.032	-.059	-.082	-.102	-.361	-.396	-.435	-.511	.839
	.900										.900
Lower surface	.072	-.465	-.389	-.345	.011	.161	.290	.334	.372	.398	.072
	.146	-.455	-.351	-.138	-.008	.101	.218	.260	.304	.344	.146
	.249	-.449	-.325	-.090	.018	.099	.182	.212	.249	.288	.249
	.334	-.457	-.336	-.065	.023	.085	.149	.166	.202	.239	.334
	.430	-.465	-.323	-.040	.031	.081	.123	.129	.161	.194	.430
	.528	-.468	-.265	-.019	.038	.078	.099	.091	.114	.145	.528
	.587	-.461	-.216	-.008	.043	.076	.081	.068	.089	.114	.587
	.683	-.442	-.139	.013	.052	.076	.050	.027	.040	.060	.683
	.785	-.403	-.075	.038	.066	.078	.012	-.019	-.010	-.001	.785
	M = 0.900; q = 354 lb/sq ft										
Upper surface	.034	.364	.272	.136	-.179	-.933	-.701	-.535	-.576	-.629	.034
	.082	.273	.173	.043	-.197	-.824	-.658	-.521	-.565	-.619	.082
	.162	.188	.093	-.020	-.217	-.539	-.628	-.510	-.555	-.612	.162
	.252	.119	.035	-.069	-.228	-.392	-.601	-.500	-.548	-.606	.252
	.347	.068	-.003	-.097	-.234	-.339	-.579	-.492	-.539	-.600	.347
	.443	.028	-.039	-.125	-.244	-.345	-.552	-.480	-.529	-.591	.443
	.544	-.015	-.075	-.150	-.252	-.323	-.533	-.478	-.527	-.588	.544
	.589	-.013	-.068	-.136	-.223	-.285	-.524	-.476	-.523	-.586	.589
	.644	-.027	-.074	-.132	-.205	-.249	-.510	-.475	-.521	-.582	.644
	.738	-.038	-.069	-.111	-.163	-.179	-.486	-.471	-.513	-.576	.738
	.839	-.046	-.046	-.066	-.104	-.100	-.457	-.469	-.506	-.568	.839
	.900										.900
Lower surface	.072	-.496	-.412	-.396	.001	.159	.279	.328	.369	.401	.072
	.146	-.487	-.366	-.157	-.020	.099	.206	.255	.307	.353	.146
	.249	-.481	-.341	-.098	.010	.097	.175	.207	.251	.299	.249
	.334	-.489	-.335	-.072	.015	.087	.144	.165	.208	.255	.334
	.430	-.500	-.319	-.046	.023	.083	.119	.129	.167	.212	.430
	.528	-.510	-.279	-.021	.034	.081	.095	.092	.125	.166	.528
	.587	-.503	-.233	-.008	.037	.079	.079	.070	.100	.137	.587
	.683	-.499	-.168	.016	.049	.078	.049	.031	.051	.086	.683
	.785	-.469	-.100	.040	.065	.082	.014	-.013	.001	.028	.785

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 367 lb/sq ft										
Upper surface	.034	.358	.279	.115	-.224	-.834	-1.012	-.624	-.671	-.717	.034
	.082	.267	.179	.025	-.265	-.783	-.769	-.610	-.645	-.688	.082
	.162	.181	.094	-.033	-.302	-.540	-.741	-.598	-.634	-.679	.162
	.252	.117	.034	-.082	-.344	-.512	-.723	-.585	-.623	-.675	.252
	.347	.070	-.006	-.107	-.371	-.518	-.703	-.574	-.614	-.666	.347
	.443	.030	-.043	-.140	-.365	-.543	-.675	-.560	-.603	-.656	.443
	.544	-.010	-.083	-.171	-.269	-.604	-.652	-.556	-.600	-.653	.544
	.589	-.008	-.075	-.156	-.197	-.603	-.637	-.551	-.596	-.649	.589
	.644	-.019	-.081	-.146	-.144	-.565	-.624	-.551	-.595	-.645	.644
	.738	-.026	-.075	-.122	-.125	-.215	-.591	-.543	-.586	-.637	.738
	.839	-.026	-.050	-.074	-.077	-.045	-.557	-.536	-.576	-.629	.839
	.900										.900
Lower surface	.072	-.568	-.431	-.394	-.011	.121	.247	.321	.376	.422	.072
	.146	-.546	-.396	-.163	-.027	.068	.175	.251	.320	.378	.146
	.249	-.526	-.363	-.102	.009	.074	.150	.206	.265	.326	.249
	.334	-.548	-.365	-.075	.018	.067	.120	.165	.224	.283	.334
	.430	-.575	-.360	-.047	.030	.070	.098	.134	.187	.241	.430
	.528	-.593	-.341	-.022	.043	.072	.079	.100	.148	.198	.528
	.587	-.588	-.314	-.010	.048	.074	.064	.078	.124	.172	.587
	.683	-.566	-.243	.014	.061	.077	.038	.043	.079	.121	.683
	.785	-.524	-.156	.040	.078	.086	.006	.002	.032	.067	.785
	M = 0.980; q = 382 lb/sq ft										
Upper surface	.034	.286	.203	.083	-.119	-.609	-1.170	-1.013	-.675	-.713	.034
	.082	.188	.097	-.022	-.196	-.521	-1.117	-.843	-.675	-.709	.082
	.162	.097	.006	-.117	-.251	-.426	-1.070	-.847	-.673	-.716	.162
	.252	.023	-.060	-.188	-.303	-.457	-1.044	-.840	-.677	-.727	.252
	.347	-.032	-.107	-.225	-.336	-.472	-1.020	-.827	-.677	-.735	.347
	.443	-.084	-.164	-.270	-.385	-.498	-1.003	-.796	-.677	-.736	.443
	.544	-.135	-.229	-.338	-.445	-.557	-.996	-.781	-.686	-.742	.544
	.589	-.092	-.237	-.344	-.448	-.565	-.977	-.770	-.688	-.742	.589
	.644	-.055	-.265	-.364	-.456	-.572	-.961	-.762	-.685	-.741	.644
	.738	-.072	-.287	-.391	-.476	-.579	-.949	-.739	-.681	-.735	.738
	.839	-.075	-.063	-.250	-.454	-.611	-.950	-.715	-.679	-.728	.839
	.900										.900
Lower surface	.072	-.849	-.802	-.588	-.200	.039	.212	.335	.407	.451	.072
	.146	-.820	-.740	-.523	-.193	-.009	.136	.269	.352	.411	.146
	.249	-.799	-.621	-.389	-.135	.001	.113	.224	.301	.359	.249
	.334	-.790	-.546	-.310	-.093	-.004	.088	.188	.260	.319	.334
	.430	-.764	-.467	-.241	-.051	-.001	.070	.159	.225	.282	.430
	.528	-.674	-.352	-.123	-.016	.003	.053	.130	.189	.241	.528
	.587	-.568	-.266	-.061	-.004	.002	.041	.112	.167	.214	.587
	.683	-.402	-.132	.007	.012	.003	.023	.078	.123	.166	.683
	.785	-.272	-.016	.059	.034	.004	.006	.043	.081	.115	.785

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF

0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.034	.318	.247	.132	-.064	-.457	-1.008	-1.168	-.804	-.638	.034
	.082	.224	.145	.032	-.142	-.385	-.972	-1.153	-.759	-.642	.082
	.162	.136	.057	-.057	-.197	-.357	-.926	-1.124	-.744	-.647	.162
	.252	.070	-.008	-.128	-.244	-.376	-.907	-1.104	-.731	-.656	.252
	.347	.014	-.057	-.168	-.276	-.391	-.885	-1.075	-.716	-.666	.347
	.443	-.045	-.103	-.208	-.316	-.425	-.876	-1.041	-.698	-.669	.443
	.544	-.105	-.175	-.272	-.380	-.483	-.885	-1.024	-.691	-.677	.544
	.589	-.114	-.179	-.279	-.383	-.489	-.878	-.987	-.685	-.677	.589
	.644	-.144	-.205	-.302	-.397	-.496	-.860	-.926	-.681	-.677	.644
	.738	-.181	-.240	-.333	-.416	-.506	-.841	-.823	-.670	-.675	.738
	.839	-.146	-.274	-.365	-.452	-.524	-.838	-.767	-.660	-.675	.839
	.900										.900
Lower surface	.072	-.804	-.750	-.526	-.159	.064	.253	.361	.459	.496	.072
	.146	-.765	-.688	-.458	-.158	.016	.180	.293	.408	.457	.146
	.249	-.719	-.561	-.324	-.108	.031	.160	.256	.356	.409	.249
	.334	-.709	-.499	-.259	-.089	.028	.134	.222	.319	.370	.334
	.430	-.680	-.442	-.216	-.059	.028	.119	.196	.285	.333	.430
	.528	-.633	-.372	-.166	-.028	.034	.103	.171	.250	.293	.528
	.587	-.552	-.323	-.129	-.013	.033	.092	.154	.229	.270	.587
	.683	-.400	-.232	-.073	-.002	.030	.073	.124	.189	.224	.683
	.785	-.282	-.137	-.027	.013	.030	.058	.094	.147	.174	.785
	M = 1.125; q = 425 lb/sq ft										
Upper surface	.034	.339	.274	.173	.011	-.254	-.860	-.992	-.594	-.509	.034
	.082	.255	.178	.079	-.071	-.276	-.802	-.971	-.569	-.511	.082
	.162	.167	.089	-.001	-.128	-.309	-.764	-.937	-.558	-.514	.162
	.252	.100	.022	-.063	-.169	-.302	-.728	-.913	-.548	-.513	.252
	.347	.054	-.016	-.102	-.198	-.312	-.719	-.896	-.534	-.513	.347
	.443	.009	-.056	-.149	-.239	-.339	-.715	-.856	-.519	-.506	.443
	.544	-.052	-.120	-.205	-.302	-.396	-.730	-.845	-.515	-.511	.544
	.589	-.056	-.129	-.211	-.306	-.400	-.734	-.845	-.509	-.509	.589
	.644	-.080	-.155	-.226	-.319	-.415	-.735	-.837	-.507	-.509	.644
	.738	-.111	-.179	-.252	-.336	-.433	-.706	-.764	-.497	-.505	.738
	.839	-.145	-.207	-.294	-.359	-.448	-.684	-.692	-.487	-.501	.839
	.900										.900
Lower surface	.072	-.699	-.655	-.442	-.125	.062	.284	.418	.541	.597	.072
	.146	-.669	-.628	-.382	-.119	.020	.215	.356	.493	.561	.146
	.249	-.616	-.472	-.291	-.093	.032	.206	.326	.449	.518	.249
	.334	-.539	-.381	-.231	-.081	.034	.188	.297	.416	.480	.334
	.430	-.467	-.341	-.192	-.058	.041	.176	.275	.386	.447	.430
	.528	-.395	-.310	-.157	-.035	.063	.168	.254	.355	.413	.528
	.587	-.350	-.279	-.129	-.011	.075	.161	.239	.338	.391	.587
	.683	-.313	-.217	-.086	.016	.076	.145	.213	.303	.349	.683
	.785	-.271	-.142	-.034	.045	.089	.134	.192	.265	.304	.785

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 443 lb/sq ft										
Upper surface	.034	.352	.292	.205	.065	-.155	-.601	-.755	-.814	-.746	.034
	.082	.272	.207	.113	-.014	-.187	-.569	-.736	-.791	-.711	.082
	.162	.183	.120	.039	-.073	-.238	-.551	-.710	-.787	-.713	.162
	.252	.116	.058	-.022	-.122	-.249	-.555	-.713	-.792	-.707	.252
	.347	.078	.015	-.060	-.143	-.256	-.547	-.697	-.772	-.699	.347
	.443	.038	-.025	-.100	-.179	-.279	-.554	-.695	-.741	-.676	.443
	.544	-.031	-.084	-.159	-.239	-.324	-.575	-.705	-.727	-.673	.544
	.589	-.042	-.090	-.161	-.250	-.335	-.582	-.708	-.714	-.665	.589
	.644	-.066	-.108	-.175	-.263	-.346	-.581	-.696	-.700	-.662	.644
	.738	-.100	-.144	-.201	-.283	-.364	-.570	-.679	-.671	-.648	.738
	.839	-.139	-.185	-.231	-.308	-.383	-.555	-.676	-.646	-.633	.839
	.900										.900
Lower surface	.072	-.620	-.614	-.476	-.173	.047	.263	.408	.518	.576	.072
	.146	-.593	-.596	-.425	-.137	.012	.197	.348	.471	.539	.146
	.249	-.570	-.544	-.305	-.092	.020	.194	.326	.430	.500	.249
	.334	-.533	-.417	-.254	-.098	.014	.176	.298	.400	.467	.334
	.430	-.516	-.305	-.197	-.080	.019	.179	.284	.376	.438	.430
	.528	-.468	-.268	-.178	-.068	.025	.185	.270	.351	.409	.528
	.587	-.378	-.247	-.163	-.056	.040	.187	.258	.335	.388	.587
	.683	-.253	-.202	-.127	-.024	.066	.183	.236	.305	.354	.683
	.785	-.197	-.155	-.072	.024	.097	.185	.225	.276	.315	.785

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 310 lb/sq ft										
Upper surface	.084	.296	.260	.148	-.061	-.373	-.392	-.326	-.364	-.430	.084
	.155	.196	.151	.049	-.117	-.320	-.361	-.321	-.363	-.426	.155
	.258	.099	.066	-.023	-.149	-.292	-.335	-.318	-.361	-.420	.258
	.348	.032	.010	-.067	-.164	-.259	-.318	-.315	-.359	-.424	.348
	.449	-.027	-.034	-.094	-.157	-.228	-.297	-.313	-.359	-.426	.449
	.550	-.067	-.056	-.099	-.142	-.197	-.279	-.311	-.359	-.421	.550
	.648	-.092	-.063	-.091	-.122	-.165	-.269	-.311	-.361	-.430	.648
	.749	-.117	-.071	-.071	-.094	-.129	-.255	-.310	-.362	-.422	.749
	.800										.800
Lower surface	.109	-.206	-.332	-.526	-.068	.103	.170	.218	.260	.296	.109
	.201	-.189	-.311	-.314	-.037	.071	.106	.142	.182	.216	.201
	.256	-.189	-.301	-.235	-.048	.037	.058	.088	.130	.171	.256
	.353	-.175	-.294	-.110	-.027	.022	.016	.036	.066	.098	.353
	.457	-.164	-.274	-.044	-.005	.011	-.020	-.018	-.026	.003	.457
	.552	-.155	-.277	.009	.026	.032	-.023	-.031	-.022	-.008	.552
	.602	-.152	-.280	.020	.033	.034	-.031	-.048	-.043	-.033	.602
	.703	-.145	-.286	.036	.047	.041	-.045	-.072	-.077	-.078	.703
	M = 0.900; q = 354 lb/sq ft										
Upper surface	.084	.309	.257	.147	-.105	-.788	-.414	-.391	-.430	-.489	.084
	.155	.211	.151	.043	-.167	-.494	-.394	-.390	-.426	-.485	.155
	.258	.109	.060	-.036	-.206	-.346	-.375	-.389	-.419	-.478	.258
	.348	.037	-.004	-.093	-.225	-.319	-.364	-.389	-.422	-.478	.348
	.449	-.034	-.056	-.132	-.214	-.233	-.353	-.389	-.423	-.477	.449
	.550	-.087	-.084	-.132	-.179	-.211	-.344	-.388	-.414	-.472	.550
	.648	-.116	-.086	-.108	-.144	-.184	-.343	-.390	-.423	-.476	.648
	.749	-.140	-.089	-.084	-.113	-.146	-.339	-.390	-.411	-.469	.749
	.800										.800
Lower surface	.109	-.226	-.350	-.526	-.068	.116	.186	.217	.263	.297	.109
	.201	-.209	-.317	-.365	-.043	.077	.117	.139	.186	.223	.201
	.256	-.210	-.308	-.285	-.056	.041	.064	.084	.134	.174	.256
	.353	-.195	-.290	-.155	-.041	.014	.011	.025	.067	.108	.353
	.457	-.187	-.262	-.049	-.020	-.001	-.044	-.043	-.035	.003	.457
	.552	-.178	-.263	.018	.019	.020	-.053	-.066	-.042	-.010	.552
	.602	-.178	-.260	.035	.028	.024	-.064	-.085	-.065	-.042	.602
	.703	-.172	-.263	.054	.045	.033	-.083	-.116	-.106	-.094	.703

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.940; q = 367 lb/sq ft										
Upper surface	.084	.337	.265	.139	-.154	-.875	-.529	-.467	-.497	-.556	.084
	.155	.240	.157	.035	-.191	-.777	-.497	-.465	-.492	-.551	.155
	.258	.139	.066	-.042	-.213	-.696	-.474	-.460	-.489	-.543	.258
	.348	.065	.000	-.110	-.244	-.575	-.450	-.458	-.485	-.540	.348
	.449	-.008	-.068	-.175	-.274	-.537	-.433	-.455	-.480	-.538	.449
	.550	-.066	-.098	-.163	-.203	-.189	-.418	-.452	-.479	-.528	.550
	.648	-.086	-.100	-.114	-.116	-.055	-.413	-.455	-.476	-.531	.648
	.749	-.132	-.098	-.085	-.089	-.060	-.401	-.453	-.477	-.520	.749
	.800										.800
Lower surface	.109	-.294	-.335	-.545	-.024	.102	.170	.223	.281	.325	.109
	.201	-.280	-.311	-.360	-.013	.065	.104	.147	.206	.258	.201
	.256	-.282	-.306	-.287	-.039	.029	.050	.095	.158	.209	.256
	.353	-.274	-.292	-.155	-.031	-.003	-.002	.035	.094	.145	.353
	.457	-.263	-.275	-.054	-.015	-.033	-.069	-.039	-.007	.045	.457
	.552	-.252	-.268	.021	.027	-.009	-.091	-.068	-.019	.025	.552
	.602	-.250	-.266	.037	.037	.007	-.106	-.092	-.045	-.007	.602
	.703	-.249	-.264	.057	.055	.037	-.123	-.128	-.093	-.064	.703
	M = 0.980; q = 382 lb/sq ft										
Upper surface	.084	.303	.188	.075	-.115	-.457	-1.062	-.609	-.581	-.641	.084
	.155	.213	.077	-.043	-.219	-.511	-1.003	-.606	-.576	-.633	.155
	.258	.117	-.014	-.133	-.292	-.517	-.952	-.604	-.564	-.624	.258
	.348	.047	-.079	-.195	-.332	-.533	-.942	-.594	-.567	-.622	.348
	.449	-.027	-.148	-.265	-.390	-.555	-.946	-.587	-.568	-.619	.449
	.550	-.092	-.167	-.348	-.461	-.582	-.905	-.589	-.547	-.605	.550
	.648	-.123	-.114	-.333	-.492	-.574	-.888	-.582	-.559	-.606	.648
	.749	-.155	-.071	.006	-.479	-.569	-.819	-.587	-.533	-.592	.749
	.800										.800
Lower surface	.109	-.421	-.407	-.586	-.221	-.004	.142	.253	.320	.362	.109
	.201	-.382	-.364	-.547	-.102	-.020	.088	.182	.247	.299	.201
	.256	-.373	-.347	-.503	-.105	-.051	.040	.132	.201	.252	.256
	.353	-.360	-.310	-.424	-.108	-.076	-.003	.076	.141	.194	.353
	.457	-.356	-.290	-.294	-.097	-.120	-.068	.005	.042	.094	.457
	.552	-.363	-.304	-.126	-.024	-.119	-.104	-.034	.028	.077	.552
	.602	-.364	-.302	-.025	-.010	-.117	-.126	-.058	-.002	.044	.602
	.703	-.355	-.274	.111	.016	-.112	-.153	-.098	-.051	-.014	.703

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 400 lb/sq ft										
Upper surface	.084	.292	.228	.120	-.058	-.338	-.924	-1.036	-.610	-.632	.084
	.155	.195	.118	.009	-.161	-.409	-.888	-1.023	-.606	-.632	.155
	.258	.092	.027	-.079	-.230	-.412	-.854	-1.006	-.602	-.629	.258
	.348	.026	-.034	-.137	-.273	-.437	-.855	-.988	-.596	-.631	.348
	.449	-.051	-.114	-.201	-.322	-.464	-.822	-.944	-.591	-.632	.449
	.550	-.140	-.205	-.288	-.394	-.489	-.785	-.886	-.586	-.625	.550
	.648	-.200	-.274	-.356	-.449	-.509	-.795	-.859	-.582	-.631	.648
	.749	-.254	-.316	-.396	-.463	-.514	-.771	-.826	-.578	-.621	.749
	.800										.800
Lower surface	.109	-.565	-.566	-.708	-.390	-.010	.183	.287	.376	.408	.109
	.201	-.527	-.523	-.710	-.143	-.002	.132	.223	.306	.349	.201
	.256	-.534	-.514	-.656	-.165	-.032	.083	.175	.263	.303	.256
	.353	-.485	-.492	-.548	-.140	-.055	.049	.125	.207	.245	.353
	.457	-.469	-.474	-.413	-.138	-.082	-.010	.060	.109	.154	.457
	.552	-.440	-.481	-.205	-.081	-.085	-.043	.022	.095	.135	.552
	.602	-.435	-.489	-.104	-.058	-.083	-.064	-.007	.065	.103	.602
	.703	-.414	-.507	-.013	-.037	-.080	-.089	-.051	.013	.045	.703
	M = 1.125; q = 425 lb/sq ft										
Upper surface	.084	.334	.268	.174	.033	-.181	-.773	-.941	-.453	-.486	.084
	.155	.246	.164	.061	-.069	-.275	-.732	-.914	-.451	-.484	.155
	.258	.155	.076	-.020	-.141	-.308	-.719	-.883	-.450	-.482	.258
	.348	.099	.021	-.074	-.182	-.328	-.708	-.864	-.445	-.478	.348
	.449	.032	-.033	-.123	-.224	-.349	-.704	-.851	-.440	-.476	.449
	.550	-.051	-.118	-.202	-.295	-.396	-.701	-.832	-.438	-.474	.550
	.648	-.121	-.188	-.274	-.352	-.410	-.669	-.820	-.434	-.474	.648
	.749	-.180	-.247	-.324	-.392	-.437	-.667	-.789	-.434	-.471	.749
	.800										.800
Lower surface	.109	-.602	-.642	-.648	-.380	-.017	.238	.372	.475	.520	.109
	.201	-.563	-.623	-.627	-.172	-.002	.195	.313	.413	.466	.201
	.256	-.558	-.610	-.576	-.137	.002	.159	.270	.374	.423	.256
	.353	-.541	-.577	-.498	-.182	-.005	.123	.226	.322	.373	.353
	.457	-.516	-.525	-.387	-.145	-.008	.084	.170	.246	.297	.457
	.552	-.512	-.505	-.273	-.084	.001	.055	.132	.222	.270	.552
	.602	-.497	-.482	-.197	-.035	.004	.034	.105	.195	.240	.602
	.703	-.467	-.449	-.082	.036	.012	.009	.065	.147	.185	.703

TABLE III.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Concluded

(f) 95-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 443 lb/sq ft										
Upper surface	.084	.338	.287	.207	.085	-.092	-.526	-.704	-.688	-.492	.084
	.155	.247	.181	.102	-.013	-.193	-.517	-.685	-.683	-.484	.155
	.258	.153	.102	.031	-.071	-.217	-.501	-.653	-.673	-.481	.258
	.348	.103	.046	-.023	-.113	-.248	-.517	-.664	-.679	-.468	.348
	.449	.057	-.000	-.073	-.156	-.278	-.538	-.665	-.683	-.456	.449
	.550	-.021	-.062	-.123	-.205	-.301	-.517	-.632	-.660	-.459	.550
	.648	-.091	-.137	-.198	-.265	-.349	-.532	-.653	-.674	-.457	.648
	.749	-.162	-.185	-.240	-.303	-.357	-.498	-.609	-.647	-.477	.749
	.800										.800
Lower surface	.109	-.705	-.716	-.656	-.431	-.065	.231	.366	.463	.517	.109
	.201	-.679	-.675	-.617	-.353	-.054	.209	.317	.406	.469	.201
	.256	-.657	-.663	-.617	-.213	-.058	.168	.270	.370	.429	.256
	.353	-.654	-.652	-.547	-.214	-.048	.152	.244	.328	.387	.353
	.457	-.566	-.600	-.458	-.228	-.038	.122	.195	.262	.320	.457
	.552	-.568	-.584	-.351	-.149	.007	.115	.179	.249	.296	.552
	.602	-.536	-.543	-.306	-.119	.024	.097	.152	.221	.269	.602
	.703	-.491	-.377	-.180	-.030	.056	.073	.112	.170	.216	.703

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE

OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY

(a) 12-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.700; q = 625 lb/sq ft										
Upper surface	.000	.130	.367	.505	.533	.497	.237	-.127	-.410	-.670	.000
	.027	.333	.221	.104	-.029	-.174	-.537	-.880	-1.745	-1.547	.027
	.079	.175	.081	-.009	-.111	-.217	-.430	-.680	-.897	-1.407	.079
	.149	.093	.013	-.061	-.142	-.226	-.397	-.602	-.727	-1.127	.149
	.243	.041	-.029	-.093	-.165	-.236	-.385	-.557	-.537	-.814	.243
	.351	-.018	-.084	-.143	-.210	-.276	-.426	-.574	-.655	-.726	.351
	.447	-.059	-.119	-.173	-.235	-.297	-.444	-.534	-.582	-.639	.447
	.546	-.065	-.119	-.166	-.222	-.276	-.398	-.426	-.583	-.592	.546
	.643	-.082	-.129	-.170	-.218	-.266	-.360	-.430	-.562	-.608	.643
	.740	-.059	-.101	-.131	-.167	-.202	-.260	-.357	-.483	-.548	.740
	.820	-.061	-.092	-.114	-.141	-.168	-.206	-.290	-.432	-.558	.820
	.903	-.036	-.053	-.064	-.081	-.095	-.115	-.182	-.329	-.518	.903
Lower surface	.025	-.481	-.311	-.135	.017	.148	.384	.575	.725	.842	.025
	.068	-.269	-.148	-.052	.040	.125	.302	.464	.605	.723	.068
	.148	-.246	-.151	-.065	.007	.078	.228	.367	.492	.590	.148
	.251	-.233	-.153	-.078	-.016	.047	.179	.297	.407	.498	.251
	.350	-.225	-.156	-.088	-.032	.024	.144	.251	.350	.436	.350
	.449	-.231	-.170	-.108	-.057	-.004	.106	.201	.291	.370	.449
	.549	-.212	-.162	-.105	-.060	-.012	.087	.170	.249	.322	.549
	.647	-.187	-.145	-.097	-.060	-.019	.066	.134	.198	.262	.647
	.741										.741
	.825	-.100	-.074	-.043	-.021	.008	.065	.104	.135	.169	.825
	.882	-.068	-.049	-.024	-.006	.016	.067	.091	.110	.131	.882
	M = 0.900; q = 714 lb/sq ft										
Upper surface	.000	.271	.427	.534	.564	.539	.364	.100	-.128	-.353	.000
	.027	.340	.234	.125	.006	-.119	-.428	-.729	-1.365	-1.442	.027
	.079	.178	.088	.003	-.090	-.183	-.368	-.562	-.767	-1.410	.079
	.149	.093	.018	-.056	-.131	-.207	-.352	-.500	-.622	-.775	.149
	.243	.039	-.032	-.095	-.160	-.221	-.341	-.477	-.603	-.692	.243
	.351	-.032	-.097	-.161	-.225	-.281	-.393	-.513	-.526	-.690	.351
	.447	-.083	-.147	-.212	-.280	-.343	-.449	-.548	-.540	-.686	.447
	.546	-.091	-.151	-.211	-.280	-.348	-.463	-.552	-.576	-.630	.546
	.643	-.114	-.171	-.233	-.314	-.390	-.517	-.580	-.614	-.654	.643
	.740	-.089	-.138	-.187	-.255	-.371	-.505	-.535	-.600	-.592	.740
	.820	-.089	-.127	-.160	-.200	-.281	-.519	-.435	-.584	-.590	.820
	.903	-.058	-.078	-.091	-.105	-.116	-.258	-.279	-.495	-.567	.903
Lower surface	.025	-.436	-.286	-.125	.017	.151	.381	.582	.741	.873	.025
	.068	-.233	-.140	-.048	.039	.130	.303	.475	.619	.748	.068
	.148	-.240	-.154	-.073	-.001	.078	.228	.376	.507	.617	.148
	.251	-.242	-.165	-.091	-.027	.043	.177	.306	.424	.528	.251
	.350	-.250	-.180	-.111	-.049	.013	.139	.256	.365	.465	.350
	.449	-.287	-.216	-.144	-.083	-.022	.095	.205	.305	.402	.449
	.549	-.305	-.224	-.151	-.092	-.034	.075	.172	.263	.357	.549
	.647	-.301	-.206	-.140	-.092	-.041	.049	.131	.210	.294	.647
	.741										.741
	.825	-.129	-.099	-.066	-.038	-.006	.054	.097	.147	.211	.825
	.882	-.086	-.065	-.041	-.022	.005	.052	.082	.122	.176	.882

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 732 lb/sq ft										
Upper surface	.000	.318	.467	.562	.599	.580	.432	.176	-.033	-.247	.000
	.027	.353	.249	.142	.033	-.080	-.380	-.648	-1.230	-1.407	.027
	.079	.188	.102	.014	-.069	-.153	-.333	-.501	-.685	-1.310	.079
	.149	.100	.026	-.048	-.115	-.187	-.321	-.449	-.556	-.669	.149
	.243	.044	-.021	-.086	-.141	-.195	-.312	-.433	-.538	-.582	.243
	.351	-.032	-.096	-.159	-.207	-.257	-.363	-.474	-.571	-.580	.351
	.447	-.093	-.159	-.220	-.273	-.317	-.415	-.501	-.578	-.675	.447
	.546	-.105	-.164	-.231	-.286	-.333	-.438	-.517	-.519	-.646	.546
	.643	-.142	-.216	-.284	-.334	-.388	-.491	-.547	-.587	-.688	.643
	.740	-.120	-.189	-.276	-.333	-.383	-.484	-.556	-.604	-.660	.740
	.820	-.118	-.167	-.279	-.354	-.409	-.508	-.575	-.578	-.647	.820
	.903	-.081	-.097	-.154	-.284	-.378	-.512	-.495	-.531	-.611	.903
Lower surface	.025	-.398	-.258	-.112	.033	.163	.404	.590	.757	.889	.025
	.068	-.205	-.124	-.038	.054	.144	.325	.484	.636	.763	.068
	.148	-.221	-.145	-.070	.008	.086	.246	.385	.525	.634	.148
	.251	-.231	-.156	-.090	-.022	.048	.191	.315	.442	.546	.251
	.350	-.242	-.178	-.117	-.050	.015	.150	.264	.384	.485	.350
	.449	-.276	-.217	-.162	-.096	-.028	.101	.211	.323	.422	.449
	.549	-.308	-.255	-.193	-.117	-.048	.074	.178	.282	.377	.549
	.647	-.340	-.282	-.212	-.123	-.060	.042	.134	.229	.318	.647
	.741										.741
	.825	-.327	-.180	-.083	-.052	-.024	.041	.098	.170	.240	.825
	.882	-.265	-.083	-.049	-.032	-.016	.030	.082	.147	.208	.882
	M = 0.980; q = 776 lb/sq ft										
Upper surface	.000	.385	.506	.586	.613	.594	.467	.246			.000
	.027	.375	.277	.176	.068	-.037	-.311	-.580			.027
	.079	.212	.131	.048	-.035	-.111	-.273	-.437			.079
	.149	.125	.056	-.015	-.082	-.145	-.264	-.393			.149
	.243	.070	.009	-.048	-.104	-.156	-.263	-.383			.243
	.351	-.010	-.067	-.120	-.170	-.216	-.312	-.423			.351
	.447	-.078	-.131	-.185	-.230	-.274	-.362	-.452			.447
	.546	-.086	-.144	-.199	-.247	-.294	-.382	-.473			.546
	.643	-.149	-.200	-.252	-.303	-.347	-.434	-.526			.643
	.740	-.147	-.199	-.256	-.302	-.346	-.429	-.530			.740
	.820	-.172	-.224	-.282	-.330	-.370	-.451	-.545			.820
	.903	-.185	-.243	-.287	-.333	-.378	-.458	-.559			.903
Lower surface	.025	-.329	-.212	-.073	.057	.180	.410	.613			.025
	.068	-.162	-.090	-.005	.077	.161	.334	.507			.068
	.148	-.183	-.110	-.039	.031	.103	.257	.408			.148
	.251	-.191	-.125	-.057	.001	.065	.202	.338			.251
	.350	-.199	-.138	-.081	-.028	.031	.159	.286			.350
	.449	-.239	-.183	-.131	-.079	-.022	.109	.231			.449
	.549	-.272	-.216	-.166	-.110	-.052	.080	.194			.549
	.647	-.308	-.263	-.204	-.153	-.088	.042	.147			.647
	.741										.741
	.825	-.304	-.254	-.198	-.142	-.063	.038	.114			.825
	.882	-.302	-.251	-.192	-.136	-.061	.027	.098			.882

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 810 lb/sq ft										
Upper surface	.000	.445	.559	.627	.647	.632	.521	.321			.000
	.027	.414	.316	.210	.108	.013	-.232	-.476			.027
	.079	.257	.175	.087	.009	-.057	-.200	-.347			.079
	.149	.174	.101	.024	-.043	-.097	-.194	-.308			.149
	.243	.120	.056	-.003	-.059	-.108	-.193	-.298			.243
	.351	.045	-.017	-.072	-.118	-.164	-.239	-.341			.351
	.447	-.021	-.079	-.132	-.174	-.214	-.287	-.369			.447
	.546	-.030	-.091	-.148	-.193	-.234	-.308	-.390			.546
	.643	-.086	-.146	-.202	-.246	-.284	-.358	-.443			.643
	.740	-.089	-.146	-.203	-.245	-.283	-.356	-.446			.740
	.820	-.115	-.169	-.229	-.271	-.304	-.373	-.458			.820
	.903	-.130	-.177	-.233	-.275	-.312	-.385	-.474			.903
Lower surface	.025	-.261	-.149	-.033	.094	.215	.451	.650			.025
	.068	-.109	-.036	.034	.115	.198	.379	.549			.068
	.148	-.119	-.062	-.005	.066	.139	.304	.453			.148
	.251	-.131	-.079	-.023	.041	.105	.248	.385			.251
	.350	-.138	-.089	-.036	.020	.076	.210	.336			.350
	.449	-.176	-.135	-.083	-.031	.026	.157	.281			.449
	.549	-.208	-.162	-.115	-.062	-.001	.127	.245			.549
	.647	-.243	-.204	-.155	-.102	-.041	.089	.202			.647
	.741										.741
	.825	-.235	-.194	-.149	-.095	-.027	.086	.171			.825
	.882	-.232	-.190	-.147	-.088	-.024	.082	.158			.882
	M = 1.125; q = 859 lb/sq ft										
Upper surface	.000	.503	.585	.629	.640	.621	.550	.422			.000
	.027	.366	.274	.181	.080	-.019	-.230	-.459			.027
	.079	.222	.138	.067	-.003	-.068	-.232	-.345			.079
	.149	.158	.088	.023	-.037	-.089	-.192	-.281			.149
	.243	.127	.071	.015	-.038	-.086	-.177	-.249			.243
	.351	.057	.002	-.049	-.097	-.139	-.222	-.280			.351
	.447	.003	-.047	-.093	-.140	-.179	-.255	-.311			.447
	.546	-.016	-.065	-.111	-.155	-.193	-.267	-.329			.546
	.643	-.063	-.113	-.155	-.196	-.234	-.312	-.376			.643
	.740	-.059	-.105	-.152	-.193	-.230	-.311	-.367			.740
	.820	-.083	-.128	-.171	-.213	-.247	-.319	-.374			.820
	.903	-.095	-.142	-.184	-.225	-.258	-.323	-.390			.903
Lower surface	.025	-.251	-.139	-.039	.057	.167	.371	.595			.025
	.068	-.125	-.059	.017	.090	.165	.322	.523			.068
	.148	-.112	-.056	-.005	.048	.122	.270	.444			.148
	.251	-.109	-.059	-.007	.043	.104	.234	.386			.251
	.350	-.110	-.061	-.020	.025	.083	.210	.347			.350
	.449	-.142	-.097	-.049	-.002	.058	.163	.298			.449
	.549	-.175	-.126	-.076	-.030	.026	.135	.269			.549
	.647	-.204	-.161	-.116	-.071	-.019	.089	.233			.647
	.741										.741
	.825	-.191	-.148	-.099	-.057	-.004	.118	.224			.825
	.882	-.184	-.142	-.095	-.053	-.002	.120	.218			.882

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(a) 12-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 889 lb/sq ft										
Upper surface	.000	.538	.609	.638	.651	.634	.572	.445			.000
	.027	.358	.273	.175	.084	-.012	-.204	-.429			.027
	.079	.209	.143	.066	.006	-.060	-.220	-.318			.079
	.149	.136	.078	.016	-.027	-.077	-.175	-.271			.149
	.243	.129	.073	.016	-.036	-.084	-.165	-.232			.243
	.351	.081	.025	-.031	-.068	-.112	-.193	-.250			.351
	.447	.031	-.013	-.062	-.105	-.147	-.216	-.274			.447
	.546	.013	-.033	-.085	-.125	-.165	-.233	-.286			.546
	.643	-.032	-.085	-.135	-.171	-.207	-.273	-.327			.643
	.740	-.039	-.076	-.120	-.154	-.191	-.257	-.329			.740
	.820	-.060	-.097	-.140	-.172	-.207	-.275	-.337			.820
	.903	-.078	-.116	-.158	-.189	-.220	-.289	-.352			.903
Lower surface	.025	-.210	-.108	-.022	.071	.171	.360	.548			.025
	.068	-.099	-.038	.021	.094	.170	.312	.482			.068
	.148	-.097	-.044	.005	.051	.112	.238	.416			.148
	.211			-.011							.211
	.251	-.096	-.052		.042	.106	.239	.382			.251
	.350	-.087	-.043	-.000	.052	.109	.227	.354			.350
	.449	-.116	-.066	-.024	.025	.078	.183	.306			.449
	.549	-.144	-.100	-.060	-.009	.044	.161	.273			.549
	.647	-.175	-.132	-.094	-.045	.012	.111	.218			.647
	.741										.741
	.825	-.158	-.129	-.094	-.048	-.001	.096	.216			.825
	.882	-.151	-.117	-.096	-.053	-.007	.096	.226			.882

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 625 lb/sq ft										
Upper surface	.000	-.017	.142	.317	.388	.245	-.606	-1.364	-1.364	-1.160	.000
	.024	.334	.241	.103	-.098	-.367	-1.401	-1.458	-1.277	-1.126	.024
	.075	.158	.062	-.047	-.186	-.343	-.620	-1.170	-1.305	-1.083	.075
	.152	.063	-.024	-.114	-.216	-.321	-.578	-1.049	-1.281	-1.023	.152
	.252	-.008	-.087	-.162	-.250	-.340	-.556	-.863	-1.207	-.995	.252
	.353	-.049	-.120	-.185	-.260	-.336	-.519	-.603	-1.056	-.939	.353
	.448	-.090	-.154	-.213	-.281	-.347	-.498	-.502	-.901	-.874	.448
	.552	-.081	-.134	-.180	-.236	-.287	-.391	-.487	-.760	-.808	.552
	.652	-.074	-.117	-.154	-.198	-.236	-.300	-.420	-.653	-.758	.652
	.759	-.050	-.082	-.106	-.135	-.163	-.201	-.319	-.569	-.723	.759
	.858	-.015	-.037	-.051	-.069	-.083	-.104	-.210	-.488	-.693	.858
	.935	.027	.015	.010	.001	-.005	-.019	-.100	-.379	-.644	.935
Lower surface	.027	-.622	-.430	-.222	-.004	.164	.389	.494	.566	.605	.027
	.077	-.532	-.247	-.099	.012	.109	.291	.411	.514	.593	.077
	.154	-.433	-.195	-.085	.004	.081	.230	.345	.445	.530	.154
	.252	-.298	-.180	-.096	-.027	.039	.172	.275	.370	.454	.252
	.347	-.247	-.171	-.100	-.038	.016	.138	.229	.316	.397	.347
	.447	-.213	-.159	-.096	-.045	.007	.110	.194	.268	.340	.447
	.559	-.178	-.136	-.085	-.043	.001	.090	.160	.219	.281	.559
	.658	-.129	-.098	-.056	-.022	.015	.088	.142	.186	.236	.658
	.756	-.084	-.062	-.029	-.003	.026	.084	.120	.145	.179	.756
	.862	-.030	-.018	.006	.021	.042	.081	.092	.082	.092	.862
	.908	-.006	.002	.021	.032	.046	.077	.073	.037	.029	.908
	M = 0.900; q = 714 lb/sq ft										
Upper surface	.000	.051	.193	.323	.408	.303	-.233	-.890	-1.302	-1.068	.000
	.024	.327	.238	.113	-.060	-.302	-1.088	-1.481	-1.352	-1.024	.024
	.075	.149	.060	-.047	-.170	-.323	-.511	-1.364	-1.257	-.987	.075
	.152	.051	-.033	-.122	-.216	-.317	-.524	-.778	-1.136	-.967	.152
	.252	-.027	-.104	-.184	-.275	-.356	-.525	-.660	-1.039	-.960	.252
	.353	-.074	-.149	-.222	-.301	-.384	-.529	-.644	-.965	-.930	.353
	.448	-.124	-.199	-.288	-.377	-.452	-.588	-.665	-.882	-.880	.448
	.552	-.114	-.177	-.242	-.342	-.442	-.580	-.636	-.788	-.825	.552
	.652	-.104	-.156	-.208	-.269	-.429	-.588	-.601	-.702	-.777	.652
	.759	-.073	-.109	-.141	-.174	-.217	-.552	-.488	-.644	-.741	.759
	.858	-.030	-.053	-.068	-.082	-.079	-.168	-.283	-.594	-.717	.858
	.935	.023	.011	.006	.000	.003	-.043	-.173	-.508	-.677	.935
Lower surface	.027	-.648	-.442	-.243	-.033	.144	.374	.504	.585	.643	.027
	.077	-.453	-.256	-.105	-.003	.112	.284	.415	.527	.619	.077
	.154	-.375	-.217	-.107	-.018	.070	.221	.349	.457	.556	.154
	.252	-.346	-.217	-.127	-.053	.023	.159	.277	.382	.481	.252
	.347	-.332	-.222	-.134	-.068	.008	.125	.228	.330	.425	.347
	.447	-.332	-.223	-.141	-.079	-.018	.094	.191	.280	.370	.447
	.559	-.290	-.190	-.125	-.073	-.021	.074	.154	.231	.315	.559
	.658	-.182	-.133	-.086	-.045	-.002	.075	.136	.198	.272	.658
	.756	-.112	-.084	-.050	-.020	.013	.071	.112	.157	.221	.756
	.862	-.042	-.027	-.006	.013	.035	.066	.076	.094	.142	.862
	.908	-.012	-.001	.014	.027	.044	.060	.048	.045	.084	.908

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 732 lb/sq ft										
Upper surface	.000	.106	.222	.339	.432	.352	-.107	-.737	-1.155	-1.155	.000
	.024	.328	.243	.123	-.033	-.250	-.971	-1.336	-1.303	-1.114	.024
	.075	.147	.067	-.037	-.148	-.287	-.522	-1.236	-1.274	-1.091	.075
	.152	.050	-.033	-.117	-.207	-.285	-.473	-.696	-1.183	-1.036	.152
	.252	-.032	-.109	-.191	-.262	-.334	-.487	-.593	-1.140	-1.026	.252
	.353	-.088	-.159	-.234	-.299	-.359	-.502	-.588	-.976	-1.000	.353
	.448	-.154	-.249	-.315	-.377	-.433	-.559	-.623	-.831	-.950	.448
	.552	-.147	-.221	-.313	-.379	-.439	-.559	-.620	-.707	-.888	.552
	.652	-.138	-.206	-.319	-.390	-.450	-.569	-.635	-.673	-.831	.652
	.759	-.094	-.138	-.264	-.370	-.433	-.562	-.608	-.644	-.781	.759
	.858	-.042	-.051	-.061	-.218	-.373	-.535	-.538	-.621	-.744	.858
	.935	.023	.020	.021	-.005	-.089	-.249	-.283	-.475	-.676	.935
Lower surface	.027	-.622	-.405	-.250	-.034	.142	.388	.512	.604	.661	.027
	.077	-.462	-.245	-.115	-.002	.103	.292	.424	.541	.636	.077
	.154	-.376	-.216	-.113	-.018	.070	.232	.355	.472	.573	.154
	.252	-.329	-.211	-.144	-.062	.017	.166	.282	.398	.499	.252
	.347	-.310	-.235	-.168	-.089	-.013	.123	.235	.347	.445	.347
	.447	-.325	-.261	-.193	-.108	-.035	.092	.196	.298	.392	.447
	.559	-.350	-.289	-.197	-.102	-.042	.066	.157	.251	.338	.559
	.658	-.341	-.264	-.120	-.065	-.021	.063	.138	.219	.299	.658
	.756	-.320	-.123	-.060	-.033	-.006	.054	.113	.181	.250	.756
	.862	-.110	-.012	-.005	.008	.014	.034	.070	.121	.177	.862
	.908	-.021	.016	.020	.026	.020	.014	.037	.075	.123	.908
	M = 0.980; q = 776 lb/sq ft										
Upper surface	.000	.158	.255	.363	.445	.368	-.009	-.609			.000
	.024	.345	.263	.153	.003	-.193	-.823	-1.208			.024
	.075	.167	.093	-.003	-.108	-.240	-.496	-1.105			.075
	.152	.070	-.002	-.084	-.162	-.244	-.418	-.687			.152
	.252	-.015	-.082	-.154	-.221	-.289	-.421	-.535			.252
	.353	-.072	-.136	-.199	-.260	-.317	-.441	-.547			.353
	.448	-.163	-.218	-.276	-.333	-.386	-.492	-.593			.448
	.552	-.156	-.222	-.287	-.344	-.395	-.495	-.595			.552
	.652	-.176	-.241	-.304	-.361	-.412	-.509	-.610			.652
	.759	-.179	-.241	-.299	-.354	-.405	-.504	-.609			.759
	.858	-.171	-.231	-.292	-.345	-.392	-.488	-.599			.858
	.935	-.133	-.174	-.222	-.260	-.294	-.371	-.493			.935
Lower surface	.027	-.580	-.372	-.211	-.016	.149	.387	.532			.027
	.077	-.387	-.206	-.071	.022	.115	.291	.442			.077
	.154	-.314	-.178	-.083	.001	.080	.237	.374			.154
	.252	-.284	-.183	-.109	-.046	.024	.170	.300			.252
	.347	-.274	-.205	-.136	-.077	-.012	.123	.251			.347
	.447	-.286	-.227	-.168	-.112	-.047	.092	.208			.447
	.559	-.316	-.263	-.205	-.145	-.074	.062	.169			.559
	.658	-.312	-.260	-.201	-.140	-.065	.056	.150			.658
	.756	-.306	-.252	-.192	-.134	-.055	.044	.124			.756
	.862	-.290	-.240	-.183	-.124	-.050	.020	.080			.862
	.908	-.280	-.229	-.175	-.120	-.057	-.005	.043			.908

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 810 lb/sq ft										
Upper surface	.000	.219	.309	.398	.478	.409	.080	-.480			.000
	.024	.388	.301	.190	.050	-.136	-.717	-1.063			.024
	.075	.216	.136	.042	-.063	-.178	-.436	-.974			.075
	.152	.121	.045	-.039	-.114	-.191	-.340	-.615			.152
	.252	.040	-.035	-.105	-.168	-.231	-.344	-.443			.252
	.353	-.015	-.082	-.146	-.202	-.259	-.366	-.459			.353
	.448	-.099	-.158	-.220	-.273	-.321	-.414	-.503			.448
	.552	-.101	-.167	-.233	-.284	-.328	-.416	-.505			.552
	.652	-.119	-.187	-.251	-.302	-.344	-.432	-.520			.652
	.759	-.124	-.182	-.247	-.296	-.340	-.429	-.518			.759
	.858	-.119	-.173	-.240	-.287	-.327	-.419	-.511			.858
	.935	-.091	-.128	-.190	-.226	-.256	-.330	-.430			.935
Lower surface	.027	-.500	-.373	-.180	.022	.185	.428	.572			.027
	.077	-.337	-.146	-.027	.061	.154	.339	.486			.077
	.154	-.257	-.130	-.048	.046	.125	.283	.419			.154
	.252	-.224	-.131	-.063	.003	.071	.218	.349			.252
	.347	-.210	-.149	-.088	-.030	.035	.173	.301			.347
	.447	-.219	-.168	-.116	-.060	.003	.140	.258			.447
	.559	-.246	-.202	-.153	-.095	-.027	.110	.222			.559
	.658	-.242	-.201	-.153	-.090	-.022	.106	.205			.658
	.756	-.236	-.192	-.145	-.084	-.018	.095	.181			.756
	.862	-.222	-.180	-.136	-.076	-.014	.077	.142			.862
	.908	-.219	-.172	-.132	-.076	-.018	.054	.108			.908
	M = 1.125; q = 859 lb/sq ft										
Upper surface	.000	.284	.351	.427	.476	.462	.222	-.242			.000
	.024	.385	.298	.196	.060	-.104	-.552	-.851			.024
	.075	.224	.149	.050	-.037	-.152	-.458	-.793			.075
	.152	.133	.065	-.009	-.084	-.153	-.317	-.709			.152
	.252	.057	-.003	-.068	-.130	-.185	-.299	-.357			.252
	.353	.010	-.046	-.102	-.159	-.214	-.319	-.385			.353
	.448	-.062	-.117	-.170	-.219	-.264	-.357	-.417			.448
	.552	-.076	-.131	-.181	-.227	-.268	-.354	-.419			.552
	.652	-.093	-.146	-.197	-.243	-.283	-.361	-.437			.652
	.759	-.084	-.136	-.182	-.228	-.270	-.360	-.432			.759
	.858	-.086	-.134	-.180	-.225	-.266	-.352	-.421			.858
	.935	-.067	-.113	-.158	-.201	-.237	-.304	-.357			.935
Lower surface	.027	-.617	-.522	-.170	.004	.172	.405	.576			.027
	.077	-.253	-.125	-.021	.064	.161	.332	.496			.077
	.154	-.171	-.093	-.021	.049	.131	.283	.430			.154
	.252	-.161	-.079	-.009	.051	.113	.225	.369			.252
	.347	-.151	-.094	-.037	.011	.071	.188	.329			.347
	.447	-.165	-.119	-.065	-.016	.039	.157	.291			.447
	.559	-.188	-.145	-.095	-.051	.006	.130	.261			.559
	.658	-.191	-.145	-.093	-.045	.014	.125	.250			.658
	.756	-.191	-.142	-.093	-.048	.009	.122	.236			.756
	.862	-.182	-.138	-.091	-.046	.009	.125	.212			.862
	.908	-.183	-.137	-.090	-.046	.008	.112	.188			.908

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(b) 25-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 889 lb/sq ft										
Upper surface	.000	.324	.397	.442	.508	.481	.304	-.074			.000
	.024	.376	.295	.189	.071	-.076	-.427	-.683			.024
	.075	.219	.136	.047	-.032	-.143	-.399	-.643			.075
	.152	.138	.074	.003	-.069	-.142	-.318	-.604			.152
	.252	.070	.011	-.054	-.101	-.156	-.269	-.320			.252
	.353	.023	-.028		-.137	-.191	-.289	-.348			.353
	.448	-.039	-.092	-.147	-.194	-.240	-.324	-.388			.448
	.552	-.057	-.108	-.163	-.208	-.250	-.322	-.382			.552
	.652	-.078	-.125	-.175	-.212	-.251	-.335	-.404			.652
	.759	-.082	-.125	-.173	-.213	-.253	-.327	-.394			.759
	.858	-.077	-.125	-.175	-.209	-.248	-.327	-.387			.858
	.935	-.063	-.107	-.154	-.193	-.231	-.298	-.338			.935
Lower surface	.027	-.614	-.439	-.144	.020	.159	.393	.568			.027
	.077	-.110	-.085	-.012	.067	.156	.330	.485			.077
	.154	-.126	-.070	-.007	.069	.145	.285	.424			.154
	.252	-.126	-.071	-.018	.047	.112	.231	.365			.252
	.347	-.127	-.078	-.031	.019	.078	.193	.322			.347
	.447	-.151	-.103	-.057	-.003	.052	.166	.286			.447
	.559	-.184	-.136	-.094	-.041	.021	.129	.249			.559
	.658	-.184	-.136	-.090	-.039	.013	.126	.235			.658
	.756	-.161	-.135	-.097	-.047	.006	.116	.238			.756
	.862	-.148	-.104	-.065	-.022	.013	.110	.237			.862
	.908	-.150	-.106	-.067	-.019	.032	.121	.222			.908

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 625 lb/sq ft										
Upper surface	.000	.280	.297	.277	.109	-.890	-1.543	-1.324	-1.089	-.923	.000
	.029	.291	.194	.046	-.184	-.481	-1.387	-1.279	-.999	-.844	.029
	.077	.167	.076	-.047	-.213	-.393	-1.233	-1.211	-.990	-.831	.077
	.151	.087	-.001	-.105	-.227	-.371	-.955	-1.159	-.947	-.817	.151
	.249	.022	-.057	-.140	-.241	-.349	-.585	-1.146	-.897	-.795	.249
	.348	-.029	-.099	-.169	-.254	-.345	-.490	-1.059	-.870	-.777	.348
	.439	-.045	-.108	-.168	-.240	-.312	-.419	-.961	-.844	-.764	.439
	.549	-.034	-.088	-.137	-.195	-.251	-.319	-.758	-.804	-.748	.549
	.646	-.053	-.094	-.132	-.178	-.218	-.264	-.706	-.762	-.728	.646
	.743	-.042	-.077	-.102	-.133	-.162	-.191	-.572	-.721	-.709	.743
	.841	-.012	-.034	-.049	-.069	-.087	-.102	-.414	-.679	-.691	.841
	.914	.018	.001	-.003	-.014	-.024	-.033	-.288	-.643	-.675	.914
Lower surface	.026	-.672	-.557	-.207	.027	.200	.392	.472	.514	.540	.026
	.077	-.602	-.395	-.119	.004	.120	.294	.400	.477	.539	.077
	.149	-.583	-.264	-.065	.021	.103	.251	.350	.424	.487	.149
	.244	-.509	-.188	-.093	-.015	.055	.180	.272	.348	.416	.244
	.345	-.315	-.153	-.085	-.022	.037	.148	.226	.293	.357	.345
	.444	-.202	-.129	-.073	-.022	.029	.123	.188	.245	.301	.444
	.542	-.140	-.095	-.049	-.007	.034	.114	.163	.206	.253	.542
	.638	-.082	-.054	-.016	.016	.050	.115	.148	.173	.208	.638
	.731	-.041	-.019	.011	.036	.062	.113	.129	.131	.152	.731
	.828	.006	.020	.043	.059	.079	.115	.106	.072	.076	.828
	.877	.022	.033	.053	.066	.081	.109	.086	.023	.019	.877
	M = 0.900; q = 714 lb/sq ft										
Upper surface	.000	.293	.306	.267	.179	-.572	-1.490	-1.335	-.893	-.973	.000
	.029	.270	.175	.036	-.166	-.452	-1.397	-1.277	-.884	-.857	.029
	.077	.142	.053	-.065	-.222	-.430	-1.226	-1.233	-.899	-.861	.077
	.151	.061	-.027	-.138	-.270	-.401	-.637	-1.160	-.913	-.855	.151
	.249	-.006	-.087	-.182	-.291	-.406	-.605	-1.077	-.887	-.831	.249
	.348	-.058	-.136	-.222	-.319	-.448	-.637	-.978	-.854	-.811	.348
	.439	-.074	-.144	-.220	-.342	-.449	-.647	-.886	-.830	-.797	.439
	.549	-.063	-.123	-.189	-.262	-.428	-.587	-.757	-.798	-.778	.549
	.646	-.077	-.123	-.169	-.225	-.324	-.653	-.714	-.765	-.759	.646
	.743	-.061	-.096	-.126	-.157	-.152	-.305	-.657	-.736	-.737	.743
	.841	-.024	-.045	-.061	-.078	-.075	-.097	-.585	-.707	-.720	.841
	.914	.012	-.000	-.007	-.015	-.013	-.031	-.509	-.683	-.706	.914
Lower surface	.026	-.717	-.604	-.237	-.020	.162	.369	.471	.533	.566	.026
	.077	-.637	-.386	-.151	-.029	.091	.269	.391	.484	.557	.077
	.149	-.602	-.317	-.102	.005	.081	.235	.345	.430	.509	.149
	.244	-.528	-.270	-.135	-.050	.029	.159	.264	.354	.440	.244
	.345	-.414	-.225	-.127	-.055	.013	.125	.219	.300	.383	.345
	.444	-.300	-.178	-.108	-.050	.007	.102	.180	.252	.330	.444
	.542	-.200	-.130	-.076	-.029	.017	.095	.154	.216	.286	.542
	.638	-.121	-.076	-.036	.001	.038	.098	.138	.184	.246	.638
	.731	-.065	-.030	.000	.028	.056	.098	.115	.145	.195	.731
	.828	-.007	.019	.040	.058	.078	.101	.086	.092	.129	.828
	.877	.013	.035	.052	.067	.083	.096	.058	.048	.080	.877

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 732 lb/sq ft										
Upper surface	.000	.302	.316	.272	.230	-.384	-1.354	-1.427	-1.009	-.889	.000
	.029	.260	.166	.034	-.139	-.399	-1.264	-1.302	-.963	-.882	.029
	.077	.131	.041	-.071	-.212	-.377	-1.131	-1.274	-.942	-.914	.077
	.151	.046	-.044	-.156	-.260	-.372	-.563	-1.121	-.905	-.938	.151
	.249	-.027	-.112	-.212	-.299	-.386	-.566	-1.038	-.880	-.915	.249
	.348	-.087	-.171	-.256	-.352	-.441	-.599	-.954	-.870	-.887	.348
	.439	-.105	-.200	-.294	-.368	-.454	-.623	-.913	-.864	-.870	.439
	.549	-.123	-.193	-.320	-.400	-.468	-.615	-.811	-.845	-.853	.549
	.646	-.101	-.152	-.297	-.406	-.486	-.634	-.749	-.814	-.830	.646
	.743	-.077	-.100	-.124	-.381	-.479	-.633	-.669	-.782	-.803	.743
	.841	-.028	-.040	-.044	-.078	-.247	-.532	-.564	-.755	-.780	.841
	.914	.017	.009	.007	.016	-.044	-.213	-.473	-.730	-.764	.914
Lower surface	.026	-.748	-.578	-.298	-.042	.147	.376	.475	.546	.584	.026
	.077	-.652	-.391	-.188	-.044	.080	.272	.392	.496	.573	.077
	.149	-.615	-.329	-.132	-.010	.076	.241	.347	.445	.525	.149
	.244	-.502	-.302	-.193	-.079	.011	.156	.265	.370	.458	.244
	.345	-.411	-.289	-.195	-.087	-.009	.118	.219	.316	.402	.345
	.444	-.385	-.302	-.163	-.075	-.015	.091	.180	.270	.351	.444
	.542	-.367	-.264	-.097	-.048	-.003	.079	.155	.234	.309	.542
	.638	-.319	-.088	-.042	-.010	.020	.077	.138	.205	.271	.638
	.731	-.151	-.009	.001	.024	.041	.072	.114	.168	.225	.731
	.828	.000	.038	.046	.062	.067	.067	.087	.120	.164	.828
	.877	.031	.051	.060	.075	.073	.056	.064	.082	.118	.877
	M = 0.980; q = 776 lb/sq ft										
Upper surface	.000	.328	.336	.292	.268	-.244	-1.134	-1.355			.000
	.029	.268	.184	.062	-.090	-.337	-1.064	-1.317			.029
	.077	.140	.061	-.044	-.168	-.316	-.967	-1.265			.077
	.151	.054	-.030	-.123	-.221	-.322	-.493	-1.197			.151
	.249	-.024	-.099	-.182	-.258	-.340	-.506	-.918			.249
	.348	-.092	-.155	-.240	-.322	-.392	-.526	-.833			.348
	.439	-.137	-.201	-.266	-.338	-.416	-.557	-.734			.439
	.549	-.169	-.237	-.305	-.370	-.427	-.550	-.664			.549
	.646	-.179	-.247	-.319	-.389	-.450	-.565	-.662			.646
	.743	-.195	-.260	-.326	-.390	-.450	-.565	-.664			.743
	.841	-.189	-.260	-.324	-.384	-.440	-.555	-.658			.841
	.914	-.107	-.162	-.218	-.263	-.319	-.420	-.593			.914
Lower surface	.026	-.714	-.543	-.291	-.033	.140	.368	.489			.026
	.077	-.598	-.351	-.156	-.039	.079	.267	.403			.077
	.149	-.540	-.288	-.101	.003	.089	.236	.359			.149
	.244	-.442	-.270	-.170	-.093	-.006	.152	.274			.244
	.345	-.363	-.261	-.188	-.120	-.044	.112	.226			.345
	.444	-.345	-.283	-.214	-.145	-.054	.084	.187			.444
	.542	-.339	-.282	-.209	-.137	-.049	.068	.160			.542
	.638	-.316	-.258	-.191	-.121	-.037	.063	.142			.638
	.731	-.298	-.237	-.172	-.101	-.023	.052	.116			.731
	.828	-.245	-.187	-.123	-.055	-.003	.037	.083			.828
	.877	-.142	-.135	-.083	-.031	-.002	.020	.055			.877

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 810 lb/sq ft										
Upper surface	.000	.376	.380	.335	.313	-.158	-1.017	-1.208			.000
	.029	.314	.226	.110	-.035	-.293	-.941	-1.172			.029
	.077	.189	.106	.002	-.111	-.242	-.846	-1.119			.077
	.151	.107	.021	-.072	-.167	-.258	-.422	-1.058			.151
	.249	.034	-.049	-.127	-.201	-.274	-.428	-.834			.249
	.348	-.031	-.105	-.188	-.261	-.321	-.446	-.733			.348
	.439	-.075	-.141	-.210	-.282	-.351	-.477	-.645			.439
	.549	-.113	-.177	-.248	-.306	-.358	-.470	-.578			.549
	.646	-.105	-.165	-.242	-.302	-.357	-.450	-.568			.646
	.743	-.144	-.197	-.269	-.327	-.383	-.491	-.579			.743
	.841	-.148	-.199	-.267	-.321	-.374	-.483	-.574			.841
	.914	-.107	-.149	-.212	-.257	-.297	-.393	-.536			.914
Lower surface	.026	-.671	-.484	-.253	.008	.181	.406	.528			.026
	.077	-.518	-.275	-.109	.005	.123	.309	.445			.077
	.149	-.450	-.225	-.061	.048	.138	.283	.401			.149
	.244	-.358	-.214	-.120	-.044	.045	.197	.321			.244
	.345	-.285	-.198	-.134	-.067	.005	.158	.275			.345
	.444	-.277	-.218	-.163	-.094	-.014	.132	.238			.444
	.542	-.270	-.219	-.160	-.088	-.010	.118	.214			.542
	.638	-.249	-.196	-.142	-.070	.001	.115	.197			.638
	.731	-.231	-.176	-.126	-.055	.012	.106	.173			.731
	.828	-.193	-.136	-.085	-.017	.035	.094	.144			.828
	.877	-.155	-.103	-.058	-.001	.037	.079	.117			.877
	M = 1.105; q = 859 lb/sq ft										
Upper surface	.000	.412	.410	.372	.348	.023	-.807	-1.012			.000
	.029	.316	.238	.138	.013	-.225	-.726	-.968			.029
	.077	.192	.120	.037	-.070	-.216	-.658	-.914			.077
	.151	.120	.046	-.035	-.118	-.196	-.578	-.867			.151
	.249	.055	-.006	-.078	-.147	-.234	-.351	-.831			.249
	.348	-.013	-.075	-.140	-.209	-.271	-.376	-.617			.348
	.439	-.043	-.104	-.166	-.240	-.307	-.427	-.543			.439
	.549	-.081	-.137	-.186	-.238	-.297	-.418	-.503			.549
	.646	-.075	-.132	-.190	-.244	-.294	-.388	-.460			.646
	.743	-.110	-.163	-.219	-.272	-.321	-.425	-.498			.743
	.841	-.115	-.164	-.219	-.270	-.320	-.419	-.490			.841
	.914	-.107	-.156	-.206	-.253	-.296	-.381	-.467			.914
Lower surface	.026	-.709	-.546	-.222	-.007	.181	.400	.543			.026
	.077	-.444	-.214	-.074	.020	.129	.310	.462			.077
	.149	-.276	-.156	-.029	.066	.156	.295	.426			.149
	.244	-.257	-.163	-.081	-.011	.065	.202	.348			.244
	.345	-.248	-.160	-.087	-.025	.036	.159	.308			.345
	.444	-.242	-.181	-.116	-.061	.002	.151	.279			.444
	.542	-.228	-.164	-.105	-.055	.010	.144	.262			.542
	.638	-.211	-.149	-.095	-.039	.023	.149	.254			.638
	.731	-.193	-.136	-.081	-.029	.036	.152	.239			.731
	.828	-.154	-.103	-.046	.003	.061	.156	.221			.828
	.877	-.132	-.080	-.028	.018	.071	.147	.200			.877

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(c) 40-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 889 lb/sq ft										
Upper surface	.000	.460	.436	.395	.374	.100	-.638	-.828			.000
	.029	.340	.266	.172	.016	-.175	-.574	-.786			.029
	.077	.219	.156	.064	-.021	-.197	-.541	-.752			.077
	.151	.145	.073	-.010	-.085	-.163	-.507	-.722			.151
	.249	.070	.009	-.055	-.123	-.204	-.399	-.695			.249
	.348	.002	-.055	-.118	-.172	-.228	-.329	-.680			.348
	.439	-.024	-.081	-.151	-.212	-.280	-.366	-.518			.439
	.549	-.059	-.106	-.161	-.215	-.273	-.381	-.447			.549
	.646	-.064	-.115	-.198	-.217	-.265	-.353	-.408			.646
	.743	-.101	-.149	-.206	-.253	-.300	-.380	-.440			.743
	.841	-.108	-.154	-.206	-.251	-.295	-.375	-.442			.841
	.914	-.103	-.145	-.194	-.232	-.271	-.353	-.424			.914
Lower surface	.026	-.694	-.565	-.192	.017	.191	.419	.551			.026
	.077	-.595	-.147	-.038	.049	.152	.328	.464			.077
	.149	-.167	-.090	-.004	.093	.171	.325	.426			.149
	.244	-.167	-.112	-.047	.017	.087	.215	.340			.244
	.345	-.172	-.121	-.062	.001	.062	.174	.293			.345
	.444	-.202	-.148	-.098	-.042	.020	.134	.266			.444
	.542	-.215	-.161	-.104	-.044	.014	.126	.251			.542
	.638	-.192	-.155	-.104	-.047	.014	.120	.267			.638
	.731	-.160	-.120	-.084	-.039	.015	.119	.263			.731
	.828	-.135	-.081	-.038	.007	.052	.167	.255			.828
	.877	-.125	-.073	-.025	.023	.070	.178	.240			.877

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.800; q = 625 lb/sq ft										
Upper surface	.000	.058	.157	.268	.111	-.697	-1.023	-.833	-.706	-.688	.000
	.026	.348	.233	.058	-.320	-.772	-1.007	-.822	-.700	-.690	.026
	.077	.216	.108	-.026	-.236	-.495	-.979	-.812	-.687	-.680	.077
	.148	.139	.039	-.072	-.227	-.410	-.954	-.802	-.678	-.677	.148
	.251	.068	-.014	-.103	-.219	-.353	-.947	-.770	-.664	-.671	.251
	.350	.021	-.051	-.127	-.224	-.325	-.915	-.734	-.652	-.664	.350
	.452										.452
	.553	-.024	-.078	-.129	-.194	-.251	-.765	-.665	-.621	-.649	.553
	.647	-.037	-.083	-.123	-.177	-.218	-.605	-.629	-.606	-.639	.647
	.752	-.041	-.077	-.105	-.142	-.169	-.404	-.593	-.587	-.626	.752
	.854	-.008	-.035	-.051	-.074	-.090	-.225	-.563	-.568	-.612	.854
	.907	.013	-.011	-.020	-.036	-.047	-.139	-.546	-.559	-.605	.907
	.925										.925
Lower surface	.040	-.608	-.429	-.166	.061	.223	.380	.437	.465	.480	.040
	.092	-.599	-.391	-.105	.046	.160	.308	.377	.424	.461	.092
	.153	-.594	-.359	-.110	.020	.113	.248	.317	.366	.414	.153
	.251	-.588	-.301	-.065	.023	.096	.208	.267	.310	.357	.251
	.343	-.564	-.207	-.057	.016	.075	.173	.220	.256	.301	.343
	.452	-.470	-.126	-.037	.020	.068	.147	.177	.202	.242	.452
	.546	-.323	-.074	-.014	.030	.070	.133	.146	.159	.192	.546
	.651	-.147	-.033	.017	.050	.081	.126	.115	.113	.137	.651
	.799	.032	.014	.048	.067	.086	.106	.039	.018	.031	.799
	.878	.074	.029	.052	.066	.078	.084	-.035	-.065	-.058	.878
	M = 0.900; q = 714 lb/sq ft										
Upper surface	.000	.048	.157	.263	.146	-.626	-1.359	-.936	-.732	-.718	.000
	.026	.326	.217	.040	-.349	-.489	-1.462	-.807	-.734	-.724	.026
	.077	.196	.090	-.054	-.284	-.491	-1.406	-.788	-.727	-.714	.077
	.148	.120	.021	-.103	-.302	-.500	-1.318	-.752	-.700	-.704	.148
	.251	.052	-.033	-.133	-.262	-.521	-1.062	-.732	-.688	-.696	.251
	.350	.005	-.072	-.162	-.280	-.418	-.773	-.716	-.679	-.691	.350
	.452										.452
	.553	-.040	-.097	-.157	-.227	-.223	-.650	-.675	-.658	-.679	.553
	.647	-.052	-.101	-.153	-.207	-.181	-.457	-.652	-.646	-.673	.647
	.752	-.056	-.093	-.127	-.163	-.091	-.336	-.621	-.630	-.664	.752
	.854	-.016	-.045	-.063	-.083	-.042	-.206	-.590	-.612	-.655	.854
	.907	.009	-.015	-.026	-.038	-.042	-.135	-.571	-.602	-.650	.907
	.925										.925
Lower surface	.040	-.695	-.489	-.224	.015	.182	.342	.422	.465	.495	.040
	.092	-.671	-.450	-.132	.016	.141	.277	.362	.424	.476	.092
	.153	-.656	-.416	-.143	-.010	.094	.217	.300	.367	.429	.153
	.251	-.652	-.361	-.095	.001	.075	.181	.253	.312	.376	.251
	.343	-.644	-.263	-.080	-.004	.060	.148	.205	.261	.322	.343
	.452	-.572	-.157	-.054	.005	.057	.127	.163	.209	.266	.452
	.546	-.412	-.090	-.027	.022	.063	.118	.135	.169	.220	.546
	.651	-.162	-.041	.009	.046	.078	.114	.103	.127	.170	.651
	.799	.081	.013	.044	.067	.086	.095	.031	.040	.072	.799
	.878	.106	.031	.050	.066	.079	.075	-.036	-.033	-.006	.878

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 732 lb/sq ft										
Upper surface	.000	.042	.142	.243	.215	-.356	-1.216	-1.059	-.799	-.822	.000
	.026	.293	.185	.011	-.291	-.852	-1.414	-.887	-.816	-.851	.026
	.077	.164	.056	-.096	-.270	-.478	-1.294	-.870	-.783	-.805	.077
	.148	.091	-.009	-.169	-.317	-.447	-1.223	-.866	-.746	-.759	.148
	.251	.024	-.060	-.210	-.344	-.465	-1.146	-.857	-.735	-.750	.251
	.350	-.020	-.092	-.237	-.393	-.502	-.848	-.819	-.726	-.743	.350
	.452										.452
	.553	-.054	-.103	-.152	-.418	-.548	-.589	-.752	-.708	-.727	.553
	.647	-.062	-.105	-.154	-.354	-.536	-.600	-.721	-.698	-.720	.647
	.752	-.061	-.094	-.133	-.086	-.392	-.625	-.681	-.684	-.711	.752
	.854	-.018	-.041	-.063	-.035	-.075	-.306	-.637	-.670	-.702	.854
	.907	.006	-.009	-.022	-.005	-.017	-.220	-.613	-.662	-.698	.907
	.925										.925
Lower surface	.040	-.734	-.577	-.329	-.029	.147	.333	.417	.473	.508	.040
	.092	-.679	-.529	-.209	-.019	.106	.265	.356	.433	.490	.092
	.153	-.667	-.491	-.210	-.042	.061	.201	.295	.377	.443	.153
	.251	-.675	-.390	-.119	-.024	.052	.166	.248	.324	.392	.251
	.343	-.660	-.292	-.096	-.023	.037	.130	.202	.274	.340	.343
	.452	-.578	-.161	-.060	-.005	.039	.107	.162	.224	.287	.452
	.546	-.425	-.089	-.025	.019	.052	.098	.135	.187	.243	.546
	.651	-.261	-.041	.013	.049	.072	.095	.106	.147	.196	.651
	.799	-.079	.013	.051	.079	.088	.076	.041	.067	.103	.799
	.878	-.005	.033	.057	.080	.086	.054	-.019	.002	.031	.878
	M = 0.980; q = 776 lb/sq ft										
Upper surface	.000	.066	.146	.243	.266	-.150	-.946	-1.286			.000
	.026	.266	.168	.036	-.203	-.658	-1.182	-1.358			.026
	.077	.132	.036	-.078	-.219	-.451	-1.089	-1.316			.077
	.148	.056	-.032	-.141	-.263	-.397	-1.035	-1.276			.148
	.251	-.021	-.112	-.211	-.312	-.408	-.972	-1.238			.251
	.350	-.084	-.166	-.269	-.358	-.446	-.733	-1.209			.350
	.452										.452
	.553	-.156	-.234	-.315	-.409	-.503	-.606	-1.113			.553
	.647	-.189	-.265	-.345	-.417	-.501	-.624	-.998			.647
	.752	-.225	-.308	-.389	-.485	-.550	-.663	-.951			.752
	.854	-.059	-.193	-.344	-.448	-.537	-.660	-.847			.854
	.907	-.016	-.044	-.129	-.236	-.390	-.638	-.738			.907
	.925										.925
Lower surface	.040	-.651	-.541	-.369	-.093	.089	.300	.409			.040
	.092	-.598	-.504	-.233	-.079	.059	.234	.351			.092
	.153	-.584	-.474	-.247	-.109	.007	.171	.287			.153
	.251	-.576	-.411	-.220	-.103	-.008	.137	.238			.251
	.343	-.580	-.398	-.234	-.135	-.027	.095	.190			.343
	.452	-.590	-.367	-.231	-.137	-.032	.067	.149			.452
	.546	-.538	-.324	-.209	-.117	-.022	.052	.121			.546
	.651	-.430	-.279	-.171	-.073	-.006	.040	.094			.651
	.799	-.245	-.127	-.041	.006	-.001	.002	.034			.799
	.878	-.154	-.039	.014	.020	-.013	-.030	-.010			.878

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 810 lb/sq ft										
Upper surface	.000	.124	.201	.287	.312	-.056	-.838	-1.144			.000
	.026	.308	.222	.090	-.146	-.579	-1.079	-1.218			.026
	.077	.178	.091	-.020	-.155	-.386	-.970	-1.172			.077
	.148	.106	.024	-.088	-.197	-.330	-.917	-1.131			.148
	.251	.033	-.060	-.150	-.245	-.344	-.863	-1.096			.251
	.350	-.033	-.110	-.207	-.291	-.373	-.730	-1.074			.350
	.452										.452
	.553	-.107	-.174	-.255	-.346	-.434	-.517	-1.029			.553
	.647	-.137	-.204	-.280	-.349	-.428	-.543	-.860			.647
	.752	-.181	-.246	-.334	-.415	-.476	-.582	-.814			.752
	.854	-.161	-.235	-.320	-.396	-.472	-.579	-.750			.854
	.907	-.107	-.178	-.277	-.360	-.431	-.571	-.664			.907
	.925										.925
Lower surface	.040	-.556	-.467	-.321	-.038	.130	.341	.449			.040
	.092	-.503	-.422	-.180	-.035	.103	.281	.392			.092
	.153	-.492	-.391	-.190	-.060	.045	.217	.332			.153
	.251	-.482	-.334	-.163	-.052	.036	.183	.285			.251
	.343	-.465	-.317	-.180	-.087	.008	.145	.240			.343
	.452	-.461	-.293	-.180	-.087	-.001	.118	.200			.452
	.546	-.433	-.261	-.160	-.071	.011	.106	.175			.546
	.651	-.376	-.226	-.127	-.043	.024	.093	.146			.651
	.799	-.269	-.142	-.059	.005	.035	.056	.087			.799
	.878	-.199	-.092	-.037	.001	.016	.022	.045			.878
	M = 1.125; q = 859 lb/sq ft										
Upper surface	.000	.186	.257	.349	.376	.073	-.627	-.949			.000
	.026	.338	.262	.140	-.074	-.477	-.878	-1.009			.026
	.077	.211	.138	.048	-.088	-.296	-.775	-.963			.077
	.148	.142	.066	-.018	-.124	-.253	-.737	-.925			.148
	.251	.053	-.011	-.084	-.171	-.265	-.711	-.895			.251
	.350	-.007	-.068	-.139	-.210	-.294	-.687	-.881			.350
	.452										.452
	.553	-.076	-.137	-.209	-.286	-.347	-.402	-.869			.553
	.647	-.107	-.160	-.221	-.287	-.360	-.430	-.860			.647
	.752	-.147	-.219	-.285	-.339	-.393	-.476	-.742			.752
	.854	-.147	-.210	-.274	-.346	-.397	-.490	-.642			.854
	.907	-.133	-.201	-.265	-.331	-.395	-.483	-.583			.907
	.925										.925
Lower surface	.040	-.474	-.424	-.233	.018	.170	.354	.484			.040
	.092	-.411	-.338	-.096	.018	.139	.301	.431			.092
	.153	-.412	-.291	-.114	-.012	.090	.234	.372			.153
	.251	-.388	-.236	-.093	-.007	.071	.200	.330			.251
	.343	-.346	-.229	-.116	-.048	.027	.168	.291			.343
	.452	-.323	-.219	-.131	-.066	.010	.149	.258			.452
	.546	-.309	-.204	-.122	-.058	.014	.150	.240			.546
	.651	-.291	-.186	-.104	-.039	.029	.146	.219			.651
	.799	-.240	-.129	-.050	.007	.064	.131	.167			.799
	.878	-.205	-.110	-.039	.008	.057	.101	.129			.878

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(d) 60-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 889 lb/sq ft										
Upper surface	.000	.195	.274	.336	.391	.143	-.458	-.737			.000
	.026	.334	.259	.142	-.042	-.372	-.718	-.848			.026
	.077	.210	.146	.050	-.068	-.259	-.635	-.790			.077
	.148	.134	.068	-.019	-.116	-.254	-.610	-.768			.148
	.251	.057	-.004	-.080	-.160	-.243	-.597	-.754			.251
	.350	-.000	-.059	-.128	-.197	-.273	-.586	-.743			.350
	.452										.452
	.553	-.061	-.119	-.191	-.248	-.304	-.496	-.732			.553
	.647	-.076	-.127	-.195	-.257	-.317	-.406	-.716			.647
	.752	-.136	-.191	-.246	-.301	-.365	-.440	-.713			.752
	.854	-.132	-.189	-.262	-.311	-.371	-.442	-.730			.854
	.907	-.127	-.184	-.251	-.311	-.364	-.439	-.544			.907
	.925										.925
Lower surface	.040	-.733	-.643	-.276	.002	.167	.339	.474			.040
	.092	-.635	-.519	-.090	.014	.128	.292	.422			.092
	.153	-.606	-.236	-.108	-.008	.088	.218	.363			.153
	.251	-.416	-.154	-.094	-.004	.072	.197	.336			.251
	.343	-.179	-.158	-.108	-.041	.027	.156	.306			.343
	.452	-.189	-.159	-.105	-.049	.011	.162	.282			.452
	.546	-.180	-.143	-.088	-.028	.031	.164	.277			.546
	.651	-.172	-.124	-.072	-.012	.053	.168	.265			.651
	.799	-.129	-.079	-.020	.041	.101	.177	.224			.799
	.878	-.116	-.063	-.003	.049	.100	.160	.187			.878

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 625 lb/sq ft										
Upper surface	.034	.354	.281	.157	-.114	-.668	-.566	-.496	-.515	-.577	.034
	.082	.261	.180	.064	-.139	-.484	-.551	-.484	-.506	-.566	.082
	.162	.192	.120	.016	-.142	-.367	-.533	-.474	-.497	-.558	.162
	.252	.116	.047	-.043	-.173	-.324	-.509	-.465	-.490	-.553	.252
	.347	.072	.012	-.066	-.176	-.296	-.478	-.457	-.483	-.549	.347
	.443	.025	-.028	-.097	-.191	-.285	-.445	-.445	-.477	-.544	.443
	.544	-.012	-.058	-.114	-.194	-.263	-.412	-.429	-.469	-.539	.544
	.589	-.013	-.050	-.100	-.172	-.234	-.394	-.421	-.465	-.537	.589
	.644	-.027	-.058	-.084	-.160	-.212	-.380	-.418	-.462	-.534	.644
	.738	-.040	-.053	-.051	-.129	-.163	-.356	-.408	-.453	-.526	.738
	.839	-.050	-.032	-.005	-.080	-.101	-.334	-.399	-.443	-.516	.839
	.900										.900
Lower surface	.072	-.486	-.589	-.235	.001	.145	.280	.319	.352	.377	.072
	.146	-.479	-.567	-.147	-.007	.100	.221	.256	.289	.323	.146
	.249	-.486	-.561	-.091	.008	.088	.172	.197	.230	.265	.249
	.334	-.497	-.293	-.062	.015	.080	.146	.158	.188	.221	.334
	.430	-.486	-.143	-.038	.023	.075	.120	.119	.144	.173	.430
	.528	-.442	-.074	-.014	.034	.074	.096	.084	.102	.126	.528
	.587	-.410	-.050	-.002	.039	.072	.079	.061	.076	.097	.587
	.683	-.352	-.014	.020	.050	.074	.051	.021	.029	.043	.683
	.785	-.286	.020	.043	.064	.078	.016	-.023	-.023	-.017	.785
	M = 0.900; q = 714 lb/sq ft										
Upper surface	.034	.351	.277	.148	-.137	-.956	-.946	-.538	-.578	-.637	.034
	.082	.259	.175	.052	-.164	-.638	-.784	-.524	-.566	-.623	.082
	.162	.190	.114	.004	-.163	-.418	-.770	-.513	-.555	-.613	.162
	.252	.113	.040	-.058	-.197	-.350	-.749	-.504	-.547	-.607	.252
	.347	.068	.003	-.085	-.206	-.343	-.703	-.499	-.541	-.603	.347
	.443	.020	-.041	-.121	-.231	-.349	-.646	-.493	-.536	-.600	.443
	.544	-.021	-.073	-.142	-.236	-.327	-.589	-.487	-.533	-.596	.544
	.589	-.021	-.067	-.130	-.211	-.289	-.562	-.484	-.531	-.594	.589
	.644	-.036	-.073	-.127	-.193	-.251	-.538	-.484	-.530	-.592	.644
	.738	-.048	-.068	-.106	-.153	-.181	-.490	-.480	-.523	-.586	.738
	.839	-.059	-.044	-.065	-.090	-.101	-.444	-.476	-.515	-.577	.839
	.900										.900
Lower surface	.072	-.497	-.605	-.364	-.013	.144	.260	.302	.345	.376	.072
	.146	-.489	-.576	-.158	-.020	.098	.202	.241	.283	.327	.146
	.249	-.486	-.566	-.107	-.002	.085	.161	.183	.227	.273	.249
	.334	-.490	-.376	-.076	.007	.079	.139	.147	.187	.231	.334
	.430	-.493	-.220	-.051	.017	.076	.116	.111	.146	.187	.430
	.528	-.481	-.121	-.025	.029	.075	.097	.077	.106	.144	.528
	.587	-.463	-.083	-.011	.035	.073	.084	.054	.081	.115	.587
	.683	-.430	-.035	.013	.047	.075	.058	.015	.035	.064	.683
	.785	-.385	.007	.039	.063	.079	.027	-.027	-.015	.007	.785

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 732 lb/sq ft										
Upper surface	.034	.340	.181	.134	-.188	-.892	-1.376	-.663	-.653	-.703	.034
	.082	.251	.119	.042	-.235	-.557	-1.282	-.611	-.633	-.674	.082
	.162	.184	.042	-.002	-.260	-.493	-1.213	-.587	-.619	-.663	.162
	.252	.109	.004	-.065	-.332	-.503	-1.168	-.572	-.610	-.656	.252
	.347	.066	-.041	-.094	-.347	-.496	-1.116	-.562	-.602	-.651	.347
	.443	.017	-.076	-.134	-.267	-.533	-.923	-.554	-.597	-.646	.443
	.544	-.022	-.070	-.161	-.176	-.593	-.661	-.547	-.592	-.641	.544
	.589	-.022	-.076	-.146	-.173	-.589	-.622	-.543	-.590	-.639	.589
	.644	-.036	-.070	-.140	-.189	-.472	-.585	-.543	-.587	-.637	.644
	.738	-.049	-.044	-.118	-.160	-.127	-.540	-.536	-.581	-.630	.738
	.839	-.056	-.011	-.071	-.082	-.030	-.505	-.528	-.573	-.622	.839
	.900										.900
Lower surface	.072	-.651	-.597	-.392	-.026	.108	.231	.296	.348	.386	.072
	.146	-.636	-.571	-.164	-.026	.069	.176	.236	.288	.338	.146
	.249	-.634	-.544	-.114	-.003	.065	.139	.181	.236	.288	.249
	.334	-.626	-.446	-.079	.011	.066	.122	.147	.199	.249	.334
	.430	-.577	-.279	-.052	.026	.069	.106	.114	.161	.207	.430
	.528	-.476	-.152	-.024	.040	.074	.094	.082	.124	.166	.528
	.587	-.421	-.107	-.011	.047	.075	.084	.062	.100	.140	.587
	.683	-.349	-.049	.015	.060	.081	.067	.026	.058	.092	.683
	.785	-.289	-.003	.043	.076	.090	.046	-.012	.013	.040	.785
	M = 0.980; q = 776 lb/sq ft										
Upper surface	.034	.266	.193	.099	-.082	-.488	-1.119	-1.329			.034
	.082	.173	.090	-.013	-.163	-.386	-1.054	-1.292			.082
	.162	.102	.025	-.084	-.204	-.406	-.999	-1.243			.162
	.252	.017	-.058	-.174	-.282	-.433	-.985	-1.217			.252
	.347	-.030	-.106	-.204	-.319	-.443	-.959	-1.189			.347
	.443	-.094	-.172	-.259	-.373	-.479	-.955	-1.147			.443
	.544	-.130	-.230	-.324	-.430	-.532	-.967	-.990			.544
	.589	-.086	-.241	-.333	-.432	-.534	-.970	-.930			.589
	.644	-.055	-.266	-.356	-.439	-.536	-.960	-.892			.644
	.738	-.071	-.180	-.383	-.466	-.556	-.892	-.837			.738
	.839	-.075	-.011	-.162	-.476	-.588	-.748	-.789			.839
	.900										.900
Lower surface	.072	-.976	-.881	-.737	-.227	-.000	.158	.264			.072
	.146	-.909	-.771	-.443	-.203	-.031	.105	.205			.146
	.249	-.815	-.631	-.354	-.161	-.033	.066	.154			.249
	.334	-.775	-.580	-.301	-.120	-.029	.047	.127			.334
	.430	-.700	-.489	-.253	-.059	-.023	.032	.098			.430
	.528	-.462	-.332	-.128	-.019	-.014	.019	.074			.528
	.587	-.387	-.231	-.062	-.006	-.011	.010	.057			.587
	.683	-.260	-.083	.009	.015	-.006	-.006	.028			.683
	.785	-.136	.020	.062	.039	-.000	-.020	-.001			.785

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 1.030; q = 810 lb/sq ft										
Upper surface	.034	.295	.237	.152	-.025	-.320	-1.002	-1.190			.034
	.082	.205	.138	.041	-.110	-.313	-.932	-1.152			.082
	.162	.138	.075	-.029	-.149	-.333	-.879	-1.104			.162
	.252	.061	-.013	-.120	-.220	-.355	-.863	-1.078			.252
	.347	.016	-.053	-.152	-.247	-.368	-.840	-1.055			.347
	.443	-.056	-.111	-.198	-.298	-.404	-.838	-1.045			.443
	.544	-.111	-.171	-.259	-.357	-.462	-.855	-1.010			.544
	.589	-.124	-.180	-.267	-.363	-.465	-.854	-.967			.589
	.644	-.152	-.206	-.292	-.377	-.472	-.854	-.881			.644
	.738	-.189	-.238	-.321	-.390	-.474	-.832	-.791			.738
	.839	-.152	-.273	-.355	-.439	-.505	-.690	-.744			.839
	.900										.900
Lower surface	.072	-.883	-.793	-.637	-.177	.030	.204	.305			.072
	.146	-.818	-.679	-.414	-.162	-.000	.151	.248			.146
	.249	-.729	-.558	-.307	-.129	-.005	.115	.200			.249
	.334	-.682	-.519	-.261	-.112	.000	.098	.175			.334
	.430	-.667	-.470	-.237	-.086	.002	.083	.149			.430
	.528	-.576	-.374	-.187	-.050	.012	.072	.126			.528
	.587	-.468	-.309	-.147	-.033	.012	.061	.111			.587
	.683	-.333	-.204	-.089	-.019	.010	.044	.085			.683
	.785	-.236	-.107	-.038	.001	.015	.032	.059			.785
	M = 1.175; q = 859 lb/sq ft										
Upper surface	.034	.307	.256	.184	.026	-.194	-.802	-.988			.034
	.082	.229	.171	.081	-.051	-.229	-.721	-.946			.082
	.162	.162	.098	.017	-.092	-.261	-.684	-.902			.162
	.252	.076	.008	-.058	-.154	-.268	-.678	-.879			.252
	.347	.040	-.025	-.096	-.179	-.280	-.666	-.860			.347
	.443	-.011	-.072	-.138	-.225	-.316	-.666	-.854			.443
	.544	-.066	-.124	-.196	-.282	-.366	-.687	-.864			.544
	.589	-.072	-.134	-.203	-.285	-.373	-.689	-.864			.589
	.644	-.095	-.159	-.218	-.290	-.386	-.696	-.861			.644
	.738	-.126	-.188	-.241	-.311	-.397	-.699	-.827			.738
	.839	-.160	-.222	-.288	-.340	-.424	-.709	-.681			.839
	.900										.900
Lower surface	.072	-.826	-.786	-.522	-.148	.037	.242	.353			.072
	.146	-.785	-.760	-.339	-.146	.003	.190	.299			.146
	.249	-.713	-.506	-.246	-.114	.010	.161	.256			.249
	.334	-.638	-.310	-.213	-.109	.019	.155	.237			.334
	.430	-.548	-.282	-.196	-.092	.019	.143	.215			.430
	.528	-.391	-.255	-.173	-.061	.040	.138	.197			.528
	.587	-.347	-.226	-.149	-.035	.057	.134	.184			.587
	.683	-.304	-.171	-.094	.002	.069	.117	.160			.683
	.785	-.225	-.097	-.033	.034	.081	.114	.143			.785

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(e) 80-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 889 lb/sq ft										
Upper surface	.034	.304	.253	.182	.065	-.139	-.616	-.808			.034
	.082	.237	.182	.095	-.009	-.167	-.559	-.765			.082
	.162	.171	.117	.033	-.054	-.233	-.544	-.735			.162
	.252	.091	.037	-.036	-.123	-.223	-.550	-.725			.252
	.347	.056	-.002	-.073	-.147	-.238	-.545	-.712			.347
	.443	.011	-.041	-.118	-.187	-.268	-.551	-.710			.443
	.544	-.040	-.098	-.168	-.241	-.306	-.578	-.722			.544
	.589	-.051	-.101	-.170	-.249	-.314	-.582	-.726			.589
	.644	-.075	-.117	-.186	-.264	-.335	-.587	-.724			.644
	.738	-.107	-.150	-.208	-.279	-.358	-.594	-.726			.738
	.839	-.144	-.195	-.241	-.304	-.381	-.609	-.734			.839
	.900										.900
Lower surface	.072	-.723	-.669	-.502	-.126	.054	.238	.366			.072
	.146	-.692	-.648	-.300	-.110	.016	.190	.310			.146
	.249	-.649	-.552	-.185	-.085	.010	.158	.271			.249
	.334	-.619	-.321	-.165	-.087	.007	.145	.258			.334
	.430	-.602	-.236	-.158	-.084	.012	.142	.244			.430
	.528	-.522	-.222	-.152	-.073	.014	.148	.232			.528
	.587	-.357	-.214	-.136	-.057	.028	.155	.224			.587
	.683	-.278	-.183	-.102	-.027	.054	.155	.203			.683
	.785	-.217	-.125	-.056	.022	.090	.157	.192			.785

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 0.800; q = 625 lb/sq ft										
Upper surface	.084	.290	.249	.057	-.042	-.317	-.396	-.331	-.374	-.434	.084
	.155	.189	.145	-.015	-.101	-.315	-.373	-.329	-.373	-.433	.155
	.258	.095	.059	-.059	-.138	-.281	-.350	-.327	-.375	-.432	.258
	.348	.030	.003	-.087	-.155	-.254	-.325	-.325	-.375	-.433	.348
	.449	-.027	-.037	-.096	-.157	-.224	-.302	-.322	-.375	-.436	.449
	.550	-.069	-.062	-.088	-.146	-.197	-.279	-.319	-.376	-.439	.550
	.648	-.092	-.064	-.071	-.124	-.168	-.261	-.317	-.376	-.441	.648
	.749	-.113	-.066	-.004	-.099	-.133	-.245	-.315	-.378	-.443	.749
	.800										.800
Lower surface	.109	-.220	-.364	-.508	-.085	.083	.163	.205	.242	.271	.109
	.201	-.207	-.345	-.410	-.034	.065	.110	.137	.165	.194	.201
	.256	-.201	-.337	-.330	-.051	.026	.056	.079	.112	.144	.256
	.353	-.188	-.324	-.147	-.031	.017	.016	.028	.052	.078	.353
	.457	-.176	-.317	-.034	-.019	.004	-.022	-.024	-.045	-.024	.457
	.552	-.165	-.319	.035	.021	.032	-.020	-.036	-.034	-.027	.552
	.602	-.160	-.324	.043	.029	.033	-.028	-.051	-.055	-.053	.602
	.703	-.151	-.330	.053	.045	.041	-.039	-.075	-.089	-.097	.703
	M = 0.900; q = 714 lb/sq ft										
Upper surface	.084	.298	.253	.153	-.062	-.422	-.469	-.398	-.440	-.497	.084
	.155	.197	.146	.050	-.123	-.394	-.452	-.398	-.437	-.493	.155
	.258	.100	.055	-.030	-.173	-.376	-.432	-.402	-.437	-.492	.258
	.348	.027	-.008	-.086	-.202	-.332	-.413	-.400	-.437	-.491	.348
	.449	-.042	-.061	-.126	-.200	-.244	-.395	-.398	-.439	-.492	.449
	.550	-.095	-.088	-.131	-.174	-.218	-.381	-.403	-.441	-.494	.550
	.648	-.122	-.087	-.109	-.140	-.181	-.367	-.399	-.443	-.494	.648
	.749	-.149	-.088	-.088	-.110	-.154	-.358	-.407	-.446	-.497	.749
	.800										.800
Lower surface	.109	-.232	-.361	-.515	-.094	.097	.179	.193	.236	.271	.109
	.201	-.219	-.341	-.450	-.042	.072	.120	.125	.159	.199	.201
	.256	-.213	-.331	-.383	-.065	.029	.061	.065	.109	.151	.256
	.353	-.202	-.317	-.234	-.044	.008	.011	.008	.045	.087	.353
	.457	-.192	-.304	-.073	-.028	-.009	-.044	-.060	-.062	-.016	.457
	.552	-.182	-.298	.040	.020	.022	-.048	-.079	-.060	-.029	.552
	.602	-.177	-.300	.062	.029	.025	-.058	-.098	-.085	-.060	.602
	.703	-.169	-.306	.074	.047	.035	-.075	-.126	-.126	-.113	.703

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station - Continued

	x/c	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/c
	M = 0.940; q = 732 lb/sq ft										
Upper surface	.084	.319	.154	.156	-.084	-.758	-.548	-.463	-.498	-.551	.084
	.155	.219	.063	.050	-.144	-.541	-.518	-.460	-.493	-.545	.155
	.258	.120	-.006	-.033	-.197	-.550	-.493	-.460	-.491	-.542	.258
	.348	.047	-.069	-.097	-.239	-.577	-.472	-.459	-.490	-.539	.348
	.449	-.028	-.103	-.162	-.275	-.368	-.453	-.457	-.491	-.537	.449
	.550	-.091	-.104	-.159	-.173	-.084	-.438	-.459	-.490	-.536	.550
	.648	-.120	-.102	-.119	-.129	-.089	-.428	-.459	-.492	-.534	.648
	.749	-.149	-.102	-.087	-.100	-.095	-.417	-.462	-.491	-.534	.749
	.800										.800
Lower surface	.109	-.316	-.342	-.551	-.066	.091	.155	.197	.250	.288	.109
	.201	-.298	-.322	-.449	-.024	.069	.101	.131	.177	.220	.201
	.256	-.291	-.313	-.381	-.059	.024	.038	.073	.129	.176	.256
	.353	-.280	-.300	-.212	-.040	-.001	-.012	.016	.069	.114	.353
	.457	-.263	-.285	-.062	-.022	-.027	-.085	-.057	-.032	.015	.457
	.552	-.249	-.272	.047	.029	.018	-.097	-.083	-.041	.000	.552
	.602	-.242	-.269	.063	.040	.031	-.111	-.106	-.069	-.032	.602
	.703	-.233	-.266	.073	.059	.053	-.128	-.138	-.114	-.087	.703
	M = 0.980; q = 776 lb/sq ft										
Upper surface	.084	.293	.185	.087	-.072	-.356	-.968	-1.229			.084
	.155	.197	.082	-.021	-.172	-.436	-.930	-1.186			.155
	.258	.105	.003	-.122	-.256	-.439	-.894	-1.123			.258
	.348	.038	-.043	-.182	-.308	-.469	-.911	-1.040			.348
	.449	-.037	-.090	-.255	-.363	-.499	-.921	-.956			.449
	.550	-.105	-.126	-.326	-.445	-.541	-.890	-.878			.550
	.648	-.132	-.098	-.250	-.474	-.555	-.889	-.822			.648
	.749	-.164	-.092	.013	-.375	-.565	-.835	-.786			.749
	.800										.800
Lower surface	.109	-.408	-.389	-.699	-.207	-.068	.070	.183			.109
	.201	-.381	-.359	-.633	-.120	-.036	.039	.124			.201
	.256	-.375	-.345	-.581	-.132	-.079	-.024	.070			.256
	.353	-.363	-.310	-.462	-.119	-.098	-.047	.022			.353
	.457	-.353	-.295	-.209	-.086	-.138	-.111	-.048			.457
	.552	-.345	-.292	.096	-.010	-.108	-.132	-.081			.552
	.602	-.341	-.291	.135	.006	-.098	-.149	-.110			.602
	.703	-.327	-.272	.139	.037	-.080	-.168	-.152			.703

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Continued

(f) 95-percent-semispan station - Continued

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.030; q = 810 lb/sq ft										
Upper surface	.084	.274	.220	.137	-.015	-.255	-.854	-1.115			.084
	.155	.174	.116	.036	-.115	-.339	-.817	-1.067			.155
	.258	.075	.022	-.069	-.189	-.350	-.792	-1.026			.258
	.348	.009	-.040	-.122	-.246	-.383	-.798	-1.011			.348
	.449	-.068	-.120	-.186	-.294	-.413	-.808	-1.006			.449
	.550	-.158	-.216	-.280	-.377	-.462	-.796	-.983			.550
	.648	-.218	-.285	-.355	-.441	-.486	-.791	-.956			.648
	.749	-.275	-.327	-.398	-.450	-.493	-.768	-.895			.749
	.800										.800
Lower surface	.109	-.536	-.563	-.895	-.301	-.076	.118	.223			.109
	.201	-.508	-.522	-.793	-.220	-.035	.090	.170			.201
	.256	-.500	-.502	-.677	-.216	-.068	.033	.120			.256
	.353	-.476	-.476	-.491	-.187	-.090	.005	.079			.353
	.457	-.450	-.455	-.309	-.179	-.118	-.051	.014			.457
	.552	-.429	-.452	-.117	-.085	-.091	-.072	-.014			.552
	.602	-.418	-.459	-.048	-.063	-.087	-.091	-.042			.602
	.703	-.396	-.487	-.008	-.045	-.075	-.111	-.083			.703
	M = 1.105; q = 869 lb/sq ft										
Upper surface	.084	.313	.264	.185	.063	-.117	-.648	-.911			.084
	.155	.222	.165	.085	-.023	-.201	-.620	-.865			.155
	.258	.127	.062	-.012	-.095	-.227	-.611	-.837			.258
	.348	.075	.013	-.064	-.148	-.266	-.615	-.829			.348
	.449	.013	-.041	-.113	-.191	-.297	-.624	-.827			.449
	.550	-.079	-.124	-.193	-.259	-.344	-.626	-.819			.550
	.648	-.151	-.199	-.266	-.331	-.385	-.619	-.814			.648
	.749	-.208	-.249	-.308	-.365	-.392	-.611	-.794			.749
	.800										.800
Lower surface	.109	-.778	-.844	-.757	-.390	-.084	.162	.285			.109
	.201	-.740	-.799	-.705	-.240	-.037	.156	.236			.201
	.256	-.718	-.777	-.659	-.256	-.063	.111	.191			.256
	.353	-.684	-.756	-.362	-.235	-.059	.079	.158			.353
	.457	-.632	-.713	-.333	-.229	-.058	.049	.109			.457
	.552	-.565	-.577	-.249	-.128	-.008	.034	.087			.552
	.602	-.507	-.441	-.180	-.067	.004	.016	.061			.602
	.703	-.441	-.243	-.077	.003	.024	-.005	.020			.703

TABLE IV.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR WING IN PRESENCE OF BODY - Concluded

(f) 95-percent-semispan station - Concluded

	x/c	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/c
	M = 1.200; q = 889 lb/sq ft										
Upper surface	.084	.322	.276	.198	.090	-.062	-.498	-.737			.084
	.155	.214	.166	.087	.003	-.179	-.487	-.704			.155
	.258	.133	.083	.021	-.064	-.183	-.485	-.678			.258
	.348	.080	.029	-.035	-.112	-.217	-.496	-.682			.348
	.449	.034	-.017	-.080	-.152	-.247	-.510	-.688			.449
	.550	-.036	-.084	-.142	-.205	-.284	-.515	-.670			.550
	.648	-.111	-.157	-.217	-.273	-.336	-.520	-.680			.648
	.749	-.166	-.204	-.265	-.310	-.349	-.508	-.653			.749
	.800										.800
Lower surface	.109	-.774	-.746	-.627	-.454	-.095	.167	.306			.109
	.201	-.739	-.698	-.605	-.168	-.045	.174	.271			.201
	.256	-.717	-.688	-.580	-.198	-.064	.133	.228			.256
	.353	-.708	-.664	-.410	-.199	-.071	.108	.195			.353
	.457	-.668	-.651	-.273	-.222	-.077	.084	.159			.457
	.552	-.666	-.617	-.213	-.154	-.017	.086	.148			.552
	.602	-.651	-.577	-.195	-.118	.007	.070	.127			.602
	.703	-.599	-.308	-.110	-.029	.046	.051	.086			.703

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE

OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING

(a) Station A

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.055	.121	.093	.068	.051	.030	-.003	-.021	-.030	-.028	.055
.166	.043	.021	.003	-.005	-.020	-.031	-.030	-.029	-.033	.166
.277	.019	-.004	-.021	-.026	-.032	-.033	-.035	-.034	-.041	.277
.367	.058	.026	-.003	-.020	-.045	-.097	-.150	-.200	-.200	.367
.387	.074	.035	-.005	-.036	-.078	-.152	-.241	-.321	-.326	.387
.415	.068	.021	-.028	-.064	-.114	-.209	-.318	-.388	-.419	.415
.443	.045	-.007	-.056	-.097	-.151	-.257	-.376	-.379	-.464	.443
.498	-.005	-.057	-.109	-.153	-.208	-.330	-.424	-.477	-.491	.498
.553	-.037	-.084	-.130	-.166	-.215	-.313	-.344	-.441	-.431	.553
.581	-.047	-.088	-.128	-.159	-.199	-.279	-.334	-.406	-.434	.581
.609	-.051	-.085	-.120	-.143	-.176	-.235	-.302	-.387	-.432	.609
.636	-.039	-.069	-.097	-.113	-.135	-.173	-.240	-.339	-.416	.636
.664	-.027	-.047	-.069	-.078	-.092	-.115	-.178	-.282	-.405	.664
.692	-.014	-.032	-.042	-.047	-.056	-.069	-.121	-.213	-.374	.692
.719	-.005	-.016	-.025	-.025	-.031	-.038	-.080	-.164	-.339	.719
.774	-.008	-.013	-.016	-.014	-.015	-.019	-.043	-.102	-.237	.774
.830	-.005	-.007	-.008	-.002	-.001	-.003	-.013	-.044	-.115	.830
.871	.007	.005	.007	.012	.014	.019	.012	-.004	-.040	.871
.954	.025	.029	.035	.043	.048	.042	.036	.032	.034	.954
M = 0.900; q = 354 lb/sq ft										
.055	.126	.100	.075	.050	.030	.006	-.010	-.016	-.013	.055
.166	.039	.017	.001	-.016	-.025	-.026	-.018	-.009	-.016	.166
.277	.014	-.009	-.023	-.034	-.035	-.022	-.009	.007	.003	.277
.367	.067	.037	.010	-.010	-.022	-.046	-.073	-.107	-.123	.367
.387	.083	.052	.010	-.024	-.053	-.103	-.163	-.227	-.260	.387
.415	.075	.030	-.014	-.059	-.099	-.176	-.262	-.341	-.367	.415
.443	.048	-.002	-.051	-.100	-.144	-.227	-.321	-.406	-.398	.443
.498	-.019	-.072	-.130	-.186	-.239	-.326	-.434	-.415	-.478	.498
.553	-.063	-.119	-.182	-.251	-.314	-.413	-.500	-.427	-.477	.553
.581	-.075	-.126	-.184	-.264	-.341	-.453	-.505	-.496	-.483	.581
.609	-.078	-.118	-.168	-.227	-.355	-.483	-.479	-.540	-.487	.609
.636	-.064	-.092	-.127	-.159	-.235	-.483	-.424	-.524	-.469	.636
.664	-.040	-.062	-.081	-.096	-.101	-.222	-.244	-.477	-.449	.664
.692	-.022	-.031	-.041	-.050	-.047	-.039	-.136	-.400	-.454	.692
.719	-.011	-.016	-.020	-.025	-.020	-.009	-.118	-.203	-.367	.719
.774	-.015	-.012	-.015	-.018	-.010	-.001	-.084	-.188	-.349	.774
.830	-.015	-.011	-.008	-.010	-.003	.003	-.033	-.094	-.221	.830
.871	-.003	.003	.005	.005	.011	.022	.007	-.038	-.126	.871
.954	.023	.033	.040	.041	.046	.049	.040	.017	-.022	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Continued

x/l	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/l
M = 0.940; q = 367 lb/sq ft										
.055	.133	.102	.078	.056	.038	.013	-.002	-.006	.001	.055
.166	.043	.017	-.001	-.012	-.020	-.023	-.013	-.002	-.001	.166
.277	.015	-.013	-.029	-.036	-.033	-.015	.003	.024	.029	.277
.367	.078	.044	.022	.007	.001	-.014	-.035	-.061	-.089	.367
.387	.097	.068	.023	-.003	-.027	-.069	-.124	-.176	-.232	.387
.415	.085	.038	-.001	-.039	-.075	-.144	-.221	-.295	-.353	.415
.443	.055	.003	-.044	-.085	-.126	-.205	-.290	-.367	-.449	.443
.498	-.020	-.079	-.135	-.175	-.219	-.296	-.392	-.459	-.359	.498
.553	-.081	-.150	-.211	-.259	-.305	-.378	-.465	-.496	-.463	.553
.581	-.094	-.173	-.241	-.290	-.340	-.414	-.506	-.404	-.520	.581
.609	-.100	-.170	-.264	-.314	-.367	-.455	-.537	-.424	-.566	.609
.636	-.084	-.128	-.254	-.320	-.374	-.460	-.528	-.464	-.572	.636
.664	-.052	-.070	-.111	-.304	-.388	-.479	-.513	-.477	-.534	.664
.692	-.019	-.028	-.026	-.053	-.201	-.460	-.429	-.494	-.492	.692
.719	-.003	-.011	-.009	.003	-.016	-.077	-.103	-.383	-.397	.719
.774	-.006	-.012	-.011	.004	.018	.013	-.094	-.260	-.317	.774
.830	-.010	-.013	-.011	.001	.016	.025	-.056	-.109	-.288	.830
.871	-.001	.001	.003	.011	.024	.037	-.009	-.063	-.122	.871
.954	.027	.033	.041	.049	.060	.066	.042	.015	-.060	.954
M = 0.980; q = 382 lb/sq ft										
.055	.143	.124	.092	.071	.052	.026	.015	.016	.027	.055
.166	.043	.028	.008	-.003	-.013	-.014	.003	.023	.025	.166
.277	.015	-.004	-.024	-.027	-.024	-.003	.026	.059	.069	.277
.367	.101	.078	.051	.037	.033	.024	.013	-.005	-.031	.367
.387	.122	.109	.064	.033	.010	-.027	-.068	-.116	-.169	.387
.415	.108	.071	.030	-.005	-.043	-.104	-.168	-.236	-.295	.415
.443	.078	.037	-.013	-.049	-.093	-.163	-.238	-.309	-.385	.443
.498	-.004	-.049	-.101	-.140	-.181	-.254	-.336	-.408	-.469	.498
.553	-.082	-.131	-.187	-.226	-.271	-.340	-.408	-.485	-.532	.553
.581	-.121	-.164	-.224	-.260	-.306	-.377	-.447	-.519	-.563	.581
.609	-.138	-.190	-.250	-.290	-.334	-.412	-.482	-.550	-.502	.609
.636	-.152	-.198	-.263	-.300	-.345	-.423	-.503	-.555	-.474	.636
.664	-.165	-.215	-.280	-.322	-.370	-.439	-.514	-.549	-.520	.664
.692	-.160	-.217	-.279	-.320	-.373	-.454	-.538	-.549	-.403	.692
.719	-.110	-.142	-.165	-.195	-.224	-.331	-.442	-.370	-.465	.719
.774	-.026	-.008	.001	-.010	-.035	-.132	-.110	-.134	-.256	.774
.830	-.016	.013	.021	.028	.023	-.027	-.120	-.191	-.265	.830
.871	.003	.026	.031	.046	.046	.033	-.068	-.197	-.253	.871
.954	.042	.062	.067	.083	.086	.100	.040	-.051	-.110	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Continued

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.055	.178	.153	.130	.107	.091	.067	.063	.055	.063	.055
.166	.086	.070	.055	.044	.036	.025	.038	.055	.063	.166
.277	.039	.016	.003	.000	-.002	-.012	.069	.107	.120	.277
.367	.145	.097	.069	.061	.077	.083	.078	.060	.035	.367
.387	.167	.155	.113	.083	.063	.035	.003	-.045	-.105	.387
.415	.157	.111	.072	.044	.006	-.041	-.097	-.161	-.220	.415
.443	.129	.075	.029	-.008	-.042	-.099	-.165	-.239	-.308	.443
.498	.053	-.005	-.051	-.092	-.125	-.184	-.267	-.336	-.390	.498
.553	-.022	-.085	-.134	-.172	-.206	-.269	-.339	-.406	-.456	.553
.581	-.054	-.114	-.166	-.204	-.237	-.307	-.376	-.437	-.489	.581
.609	-.079	-.140	-.191	-.234	-.266	-.339	-.412	-.476	-.519	.609
.636	-.091	-.148	-.202	-.242	-.277	-.353	-.427	-.496	-.533	.636
.664	-.108	-.161	-.221	-.263	-.295	-.371	-.441	-.510	-.561	.664
.692	-.112	-.164	-.223	-.269	-.305	-.391	-.463	-.521	-.575	.692
.719	-.091	-.114	-.159	-.183	-.201	-.307	-.437	-.540	-.500	.719
.774	-.058	-.056	-.044	-.052	-.063	-.111	-.137	-.049	-.122	.774
.830	-.068	-.060	-.051	-.047	-.058	-.082	-.077	-.102	-.202	.830
.871	-.073	-.060	-.058	-.052	-.054	-.056	-.036	-.104	-.168	.871
.954	-.126	-.104	-.061	-.039	-.033	-.022	-.049	-.154	-.200	.954
M = 1.125; q = 425 lb/sq ft										
.055	.155	.129	.101	.077	.061	.029	.006	-.003	-.008	.055
.166	.058	.050	.033	.013	.001	-.015	-.022	-.024	-.048	.166
.277	.019	.004	-.016	-.026	-.031	-.033	-.032	-.037	-.087	.277
.367	-.017	-.015	-.008	-.014	-.036	-.045	-.031	.076	.144	.367
.387	.056	.036	.016	.011	.015	.013	.008	.025	.041	.387
.415	.193	.166	.089	.036	.015	-.037	-.062	-.080	-.083	.415
.443	.144	.103	.050	.014	-.026	-.095	-.126	-.157	-.176	.443
.498	.076	.027	-.013	-.056	-.092	-.160	-.210	-.239	-.270	.498
.553	.002	-.038	-.084	-.124	-.156	-.220	-.265	-.296	-.340	.553
.581	-.025	-.068	-.113	-.155	-.191	-.255	-.303	-.338	-.370	.581
.609	-.044	-.095	-.145	-.186	-.220	-.281	-.328	-.363	-.396	.609
.636	-.055	-.100	-.151	-.192	-.225	-.291	-.343	-.378	-.403	.636
.664	-.070	-.109	-.159	-.200	-.235	-.301	-.361	-.395	-.419	.664
.692	-.072	-.126	-.178	-.215	-.250	-.316	-.371	-.406	-.428	.692
.719	-.074	-.110	-.152	-.189	-.219	-.284	-.376	-.425	-.186	.719
.774	-.044	-.057	-.064	-.066	-.071	-.079	-.086	.052	-.036	.774
.830	-.051	-.051	-.055	-.052	-.047	-.045	-.038	.012	.024	.830
.871	-.041	-.041	-.042	-.036	-.030	-.020	.016	.014	.004	.871
.954	-.076	-.053	-.037	-.018	.009	.022	.080	-.036	-.065	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.200; q = 443 lb/sq ft										
.055	.162	.139	.116	.093	.075	.047	.020	.013	.005	.055
.166	.067	.054	.039	.024	.012	-.007	-.004	-.005	-.030	.166
.277	.022	.008	-.004	-.008	-.007	-.016	-.019	-.041	-.083	.277
.367	-.013	-.016	-.011	-.016	-.034	-.041	-.053	-.068	-.062	.367
.387	.023	.022	.016	.009	.005	.006	-.010	.046	.035	.387
.415	.128	.076	.040	.038	.030	-.010	-.051	-.067	-.060	.415
.443	.164	.123	.082	.034	-.006	-.069	-.108	-.137	-.149	.443
.498	.102	.054	.013	-.027	-.063	-.135	-.180	-.215	-.230	.498
.553	.029	-.012	-.054	-.089	-.125	-.186	-.234	-.271	-.289	.553
.581	-.004	-.046	-.083	-.123	-.159	-.213	-.268	-.315	-.327	.581
.609	-.028	-.069	-.114	-.152	-.182	-.244	-.290	-.328	-.361	.609
.636	-.040	-.087	-.125	-.157	-.199	-.257	-.305	-.352	-.378	.636
.664	-.044	-.084	-.131	-.171	-.207	-.264	-.317	-.371	-.393	.664
.692	-.075	-.110	-.150	-.187	-.218	-.277	-.331	-.378	-.400	.692
.719	-.086	-.124	-.159	-.192	-.227	-.283	-.353	-.396	-.417	.719
.774	-.058	-.065	-.076	-.079	-.084	-.080	-.088	-.079	-.069	.774
.830	-.056	-.060	-.059	-.051	-.049	-.033	-.028	-.027	.047	.830
.871	-.050	-.044	-.039	-.032	-.030	-.015	-.010	-.010	.023	.871
.954	-.088	-.076	-.062	-.046	-.030	-.024	-.035	-.047	-.114	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(b) Station B

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.166	.025	.012	.003	-.005	-.018	-.051	-.067	-.096	-.107	.166
.277	.001	-.009	-.019	-.025	-.035	-.059	-.079	-.080	-.095	.277
.367	.091	.052	.013	-.013	-.041	-.124	-.215	-.272	-.284	.367
.387	.108	.066	.028	-.022	-.090	-.207	-.335	-.450	-.484	.387
.443	.058	-.007	-.064	-.120	-.185	-.311	-.457	-.488	-.675	.443
.498	-.011	-.066	-.123	-.170	-.234	-.370	-.501	-.579	-.637	.498
.553	-.054	-.107	-.157	-.190	-.252	-.358	-.418	-.557	-.636	.553
.609	-.061	-.094	-.130	-.157	-.189	-.255	-.351	-.478	-.653	.609
.664	-.035	-.055	-.070	-.081	-.094	-.119	-.202	-.376	-.661	.664
.719	-.007	-.013	-.019	-.019	-.019	-.026	-.079	-.226	-.398	.719
.774	-.018	-.023	-.024	-.023	-.018	-.017	-.044	-.133	-.216	.774
.830	-.014	-.014	-.014	-.009	-.007	-.006	-.016	-.042	-.099	.830
.871	-.004	-.004	-.003	.005	.008	.004	.005	-.002	-.043	.871
M = 0.900; q = 354 lb/sq ft										
.166	.020	.013	.001	-.013	-.024	-.044	-.056	-.081	-.096	.166
.277	-.004	-.014	-.022	-.032	-.039	-.046	-.052	-.046	-.051	.277
.367	.097	.070	.029	.002	-.011	-.065	-.121	-.154	-.190	.367
.387	.123	.082	.056	.002	-.060	-.151	-.242	-.327	-.409	.387
.443	.055	-.003	-.061	-.115	-.175	-.276	-.399	-.527	-.522	.443
.498	-.032	-.082	-.143	-.203	-.263	-.363	-.480	-.434	-.598	.498
.553	-.094	-.143	-.208	-.278	-.341	-.448	-.558	-.527	-.639	.553
.609	-.089	-.127	-.176	-.244	-.370	-.502	-.507	-.626	-.669	.609
.664	-.056	-.066	-.085	-.098	-.101	-.200	-.267	-.534	-.701	.664
.719	-.013	-.011	-.014	-.017	-.007	-.001	-.117	-.254	-.451	.719
.774	-.028	-.021	-.023	-.024	-.014	-.001	-.086	-.181	-.313	.774
.830	-.023	-.016	-.015	-.016	-.008	-.001	-.037	-.093	-.207	.830
.871	-.014	-.004	-.004	-.005	.003	.007	-.006	-.040	-.134	.871
M = 0.940; q = 367 lb/sq ft										
.166	.023	.009	-.001	-.010	-.021	-.039	-.052	-.071	-.073	.166
.277	-.003	-.018	-.028	-.032	-.038	-.040	-.040	-.026	-.030	.277
.367	.114	.082	.035	.020	.011	-.030	-.078	-.103	-.150	.367
.387	.139	.094	.076	.030	-.034	-.114	-.196	-.271	-.368	.387
.443	.064	.003	-.052	-.098	-.151	-.251	-.365	-.475	-.584	.443
.498	-.026	-.089	-.144	-.190	-.240	-.330	-.448	-.534	-.462	.498
.553	-.103	-.172	-.231	-.283	-.329	-.417	-.534	-.555	-.632	.553
.609	-.112	-.180	-.272	-.328	-.380	-.471	-.589	-.516	-.732	.609
.664	-.065	-.074	-.106	-.297	-.393	-.504	-.509	-.564	-.724	.664
.719	-.002	-.005	.002	.012	-.009	-.088	-.132	-.324	-.354	.719
.774	-.016	-.018	-.017	.000	.014	.006	-.110	-.251	-.279	.774
.830	-.018	-.017	-.014	-.003	.010	.015	-.068	-.135	-.264	.830
.871	-.011	-.008	-.005	.003	.015	.024	-.024	-.073	-.169	.871
M = 0.980; q = 382 lb/sq ft										
.166	.027	.019	.004	.001	-.015	-.033	-.036	-.047	-.047	.166
.277	-.001	-.008	-.025	-.024	-.032	-.028	-.017	.006	.010	.277
.367	.137	.119	.057	.045	.044	.007	-.028	-.046	-.087	.367
.387	.166	.132	.109	.065	-.001	-.068	-.140	-.206	-.299	.387
.443	.083	.038	-.024	-.066	-.123	-.211	-.310	-.409	-.508	.443
.498	-.013	-.055	-.116	-.151	-.207	-.285	-.386	-.480	-.570	.498
.553	-.102	-.148	-.212	-.250	-.300	-.381	-.482	-.560	-.665	.553
.609	-.150	-.197	-.266	-.304	-.357	-.433	-.534	-.619	-.705	.609
.664	-.181	-.237	-.302	-.344	-.398	-.472	-.571	-.634	-.441	.664
.719	-.105	-.127	-.141	-.159	-.184	-.247	-.281	-.250	-.354	.719
.774	-.037	-.015	-.007	-.018	-.049	-.146	-.157	-.167	-.230	.774
.830	-.020	.010	.017	.022	.010	-.052	-.149	-.192	-.258	.830
.871	-.004	.023	.023	.035	.034	.012	-.105	-.204	-.267	.871

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(b) Station B - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.166	.074	.063	.053	.047	.035	.005	-.018	-.015	-.016	.166
.277	.021	.013	.004	.004	-.008	-.035	.022	.052	.061	.277
.367	.178	.136	.060	.051	.089	.069	.037	.022	-.014	.367
.387	.213	.173	.179	.137	.056	-.002	-.070	-.131	-.222	.387
.443	.133	.074	.018	-.024	-.070	-.146	-.243	-.329	-.426	.443
.498	.044	-.013	-.061	-.101	-.143	-.216	-.310	-.400	-.490	.498
.553	-.039	-.103	-.153	-.196	-.233	-.310	-.409	-.485	-.589	.553
.609	-.089	-.149	-.203	-.244	-.282	-.366	-.461	-.539	-.657	.609
.664	-.126	-.180	-.239	-.283	-.321	-.407	-.502	-.588	-.697	.664
.719	-.086	-.097	-.123	-.134	-.146	-.205	-.269	-.316	-.224	.719
.774	-.074	-.063	-.055	-.061	-.073	-.125	-.189	-.092	-.091	.774
.830	-.072	-.063	-.056	-.052	-.065	-.108	-.118	-.128	-.223	.830
.871	-.087	-.069	-.066	-.061	-.069	-.083	-.077	-.129	-.187	.871
M = 1.125; q = 425 lb/sq ft										
.166	.039	.043	.025	.016	.002	-.024	-.074	-.094	-.138	.166
.277	.007	.000	-.010	-.022	-.029	-.047	-.080	-.098	-.138	.277
.367	-.014	.014	.015	-.002	-.022	-.033	-.023	.014	.094	.367
.387	.193	.130	.024	.006	.027	-.029	-.066	-.079	-.089	.387
.443	.149	.095	.045	-.008	-.055	-.133	-.199	-.252	-.307	.443
.498	.070	.024	-.025	-.069	-.113	-.190	-.263	-.313	-.356	.498
.553	-.019	-.060	-.108	-.155	-.188	-.262	-.346	-.390	-.446	.553
.609	-.053	-.104	-.157	-.200	-.234	-.305	-.378	-.448	-.509	.609
.664	-.093	-.133	-.188	-.228	-.262	-.337	-.416	-.502	-.533	.664
.719	-.062	-.086	-.106	-.121	-.132	-.156	-.188	-.169	-.108	.719
.774	-.052	-.063	-.067	-.074	-.077	-.096	-.128	.016	-.015	.774
.830	-.054	-.055	-.057	-.058	-.057	-.074	-.069	-.010	-.007	.830
.871	-.048	-.046	-.050	-.050	-.045	-.054	-.025	-.014	-.014	.871
M = 1.200; q = 443 lb/sq ft										
.166	.050	.043	.037	.026	.007	-.018	-.056	-.082	-.121	.166
.277	.007	.003	.001	-.008	-.014	-.031	-.053	-.089	-.125	.277
.367	-.022	.012	.016	.004	-.033	-.047	-.055	-.059	-.045	.367
.387	.175	.064	.026	.012	.043	-.014	-.068	-.108	-.128	.387
.443	.158	.108	.061	.013	-.032	-.111	-.182	-.248	-.291	.443
.498	.098	.058	.011	-.037	-.082	-.153	-.228	-.293	-.360	.498
.553	.012	-.035	-.080	-.117	-.156	-.226	-.297	-.359	-.442	.553
.609	-.036	-.078	-.112	-.147	-.187	-.261	-.341	-.422	-.511	.609
.664	-.078	-.120	-.160	-.199	-.230	-.299	-.370	-.460	-.561	.664
.719	-.078	-.091	-.106	-.122	-.136	-.149	-.176	-.207	-.211	.719
.774	-.073	-.074	-.080	-.083	-.090	-.112	-.161	-.139	-.165	.774
.830	-.070	-.067	-.065	-.065	-.062	-.074	-.082	-.056	-.010	.830
.871	-.051	-.049	-.047	-.046	-.047	-.049	-.053	-.042	-.012	.871

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(c) Station C

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.055										.055
.166	-.008	-.002	.000	.001	-.008	-.049	-.108	-.193	-.302	.166
.277	-.020	-.019	-.017	-.016	-.022	-.063	-.122	-.198	-.290	.277
.353										.353
.367	.307	.382	.389	.373	.274	.228	.140	-.039	-.214	.367
.692	.035	.037	.036	.041	.039	.026	-.051	-.071	-.258	.692
.719	.003	.005	.004	.010	.009	.007	-.027	-.090	-.278	.719
.774	-.001	-.002	-.001	.006	.005	.007	-.018	-.070	-.202	.774
.830	-.009	-.007	-.009	-.003	-.004	-.006	-.023	-.044	-.121	.830
.871	-.010	-.009	-.011	-.009	-.009	-.011	-.020	-.023	-.069	.871
.954										.954
M = 0.900; q = 354 lb/sq ft										
.055										.055
.166	-.015	-.004	-.003	-.008	-.015	-.041	-.099	-.181	-.277	.166
.277	-.027	-.020	-.020	-.028	-.028	-.054	-.103	-.167	-.246	.277
.353										.353
.367	.332	.387	.428	.399	.294	.281	.241	.107	-.049	.367
.692	.029	.038	.042	.039	.041	-.001	-.118	-.181	-.238	.692
.719	.003	.009	.013	.011	.021	.022	-.077	-.193	-.317	.719
.774	-.008	-.003	.001	-.001	.010	.015	-.075	-.189	-.333	.774
.830	-.020	-.013	-.010	-.012	-.005	-.005	-.053	-.123	-.270	.830
.871	-.023	-.017	-.016	-.016	-.012	-.011	-.036	-.073	-.168	.871
.954										.954
M = 0.940; q = 367 lb/sq ft										
.055										.055
.166	-.009	-.006	-.004	-.002	-.011	-.037	-.096	-.168	-.266	.166
.277	-.022	-.024	-.025	-.025	-.028	-.048	-.095	-.148	-.224	.277
.353										.353
.367	.348	.419	.450	.423	.310	.309	.277	.156	.008	.367
.692	.028	.044	.047	.020	-.044	-.139	-.216	-.219	-.208	.692
.719	.020	.021	.026	.032	.007	-.066	-.113	-.209	-.266	.719
.774	.006	.003	.006	.020	.025	.005	-.120	-.260	-.330	.774
.830	-.012	-.012	-.007	.000	.008	-.003	-.104	-.231	-.350	.830
.871	-.018	-.017	-.015	-.011	-.004	-.008	-.068	-.137	-.266	.871
.954										.954
M = 0.980; q = 382 lb/sq ft										
.055										.055
.166	-.004	.009	.002	.005	-.006	-.029	-.081	-.145	-.235	.166
.277	-.019	-.011	-.021	-.019	-.026	-.041	-.070	-.114	-.185	.277
.353										.353
.367	.373	.450	.481	.429	.330	.331	.315	.208	.068	.367
.692	-.116	-.107	-.118	-.116	-.132	-.174	-.247	-.244	-.145	.692
.719	-.099	-.101	-.094	-.102	-.126	-.179	-.190	-.113	-.212	.719
.774	-.020	-.002	.007	-.009	-.045	-.140	-.135	-.163	-.285	.774
.830	-.008	.019	.017	.016	-.003	-.077	-.185	-.249	-.356	.830
.871	-.005	.015	.012	.013	.007	-.030	-.179	-.277	-.384	.871
.954										.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(c) Station C - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.055										.055
.166	.043	.050	.050	.055	.044	.008	-.058	-.109	-.194	.166
.277	.001	.008	.009	.013	.004	-.039	-.039	-.072	-.133	.277
.353										.353
.367	.395	.488	.516	.494	.358	.375	.368	.265	.129	.367
.692	-.069	-.062	-.077	-.075	-.080	-.123	-.203	-.221	-.138	.692
.719	-.082	-.072	-.085	-.075	-.084	-.131	-.169	-.140	-.076	.719
.774	-.060	-.044	-.039	-.048	-.062	-.109	-.151	-.092	-.115	.774
.830	-.071	-.060	-.054	-.056	-.073	-.119	-.149	-.214	-.326	.830
.871	-.098	-.078	-.079	-.078	-.091	-.122	-.139	-.243	-.320	.871
.954										.954
M = 1.125; q = 425 lb/sq ft										
.055										.055
.166	.007	.024	.025	.021	.012	-.021	-.089	-.186	.004	.166
.277	-.007	-.008	-.006	-.011	-.016	-.047	-.125	-.216	-.305	.277
.353										.353
.367	.351	.362	.335	.321	.293	.273	.246	.033	.004	.367
.692	-.047	-.044	-.050	-.056	-.061	-.085	-.140	-.124	.115	.692
.719	-.060	-.057	-.058	-.064	-.063	-.077	-.097	-.056	.000	.719
.774	-.041	-.040	-.047	-.049	-.054	-.071	-.090	-.023	.065	.774
.830	-.049	-.049	-.054	-.060	-.066	-.083	-.107	-.105	-.044	.830
.871	-.058	-.061	-.066	-.071	-.072	-.090	-.088	-.101	-.156	.871
.954										.954
M = 1.200; q = 443 lb/sq ft										
.055										.055
.166	.020	.031	.035	.031	.021	-.017	-.075	-.170	-.283	.166
.277	-.010	.001	.001	-.001	-.005	-.033	-.107	-.203	-.348	.277
.353										.353
.367	.348	.300	.289	.282	.288	.256	.223	.065	-.211	.367
.692	-.052	-.047	-.051	-.055	-.062	-.075	-.119	-.137	-.112	.692
.719	-.069	-.061	-.060	-.059	-.063	-.061	-.076	-.086	-.103	.719
.774	-.057	-.051	-.050	-.051	-.057	-.071	-.092	-.109	-.149	.774
.830	-.066	-.059	-.062	-.066	-.073	-.087	-.129	-.162	-.196	.830
.871	-.061	-.056	-.058	-.063	-.071	-.094	-.137	-.153	-.130	.871
.954										.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(d) Station D

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.166	-.017	-.003	.007	.021	.048	.050	.063	.061	.051	.166
.277										.277
.367	-.071	-.036	-.011	.025	.049	.094	.138	.184	.218	.367
.387	-.139	-.076	-.020	.021	.059	.133	.207	.278	.336	.387
.443	-.194	-.120	-.062	-.005	.045	.150	.255	.351	.435	.443
.498	-.202	-.137	-.083	-.029	.018	.119	.211	.301	.382	.498
.553	-.131	-.080	-.036	.006	.047	.129	.203	.274	.339	.553
.609	-.128	-.097	-.062	-.028	.003	.065	.117	.163	.209	.609
.664	-.057	-.043	-.025	-.003	.012	.049	.061	.071	.074	.664
.719	-.010	-.004	.002	.011	.016	.029	.011	-.036	-.125	.719
.774	-.007	-.004	-.003	.005	.006	.013	-.007	-.060	-.181	.774
.830	-.002	-.002	-.001	.002	-.001	.003	-.013	-.044	-.143	.830
.871	-.002	-.005	-.007	-.005	-.007	-.010	-.022	-.044	-.121	.871
M = 0.900; q = 354 lb/sq ft										
.166	-.023	-.006	.003	.011	.045	.058	.073	.078	.073	.166
.277										.277
.367	-.046	-.019	.003	.026	.054	.107	.161	.210	.257	.367
.387	-.127	-.064	-.009	.023	.066	.154	.237	.313	.378	.387
.443	-.209	-.134	-.071	-.016	.041	.160	.270	.377	.472	.443
.498	-.251	-.175	-.107	-.056	.003	.120	.221	.319	.417	.498
.553	-.233	-.144	-.074	-.026	.026	.125	.206	.288	.374	.553
.609	-.194	-.133	-.087	-.053	-.011	.060	.116	.177	.254	.609
.664	-.074	-.051	-.029	-.014	.009	.039	.046	.071	.127	.664
.719	-.015	-.003	.005	.011	.022	.028	-.038	-.107	-.126	.719
.774	-.016	-.007	-.001	.001	.010	.015	-.072	-.190	-.282	.774
.830	-.013	-.007	-.004	-.005	.002	.003	-.053	-.144	-.308	.830
.871	-.015	-.011	-.007	-.012	-.007	-.009	-.046	-.111	-.221	.871
M = 0.940; q = 367 lb/sq ft										
.166	-.015	-.006	.003	.014	.047	.061	.075	.089	.084	.166
.277										.277
.367	-.016	-.003	.010	.039	.064	.118	.174	.231	.279	.367
.387	-.091	-.043	.004	.041	.078	.169	.254	.336	.401	.387
.443	-.186	-.123	-.070	-.013	.043	.169	.285	.394	.494	.443
.498	-.223	-.172	-.121	-.060	-.003	.118	.231	.340	.440	.498
.553	-.223	-.184	-.115	-.049	.008	.118	.213	.310	.399	.553
.609	-.318	-.258	-.125	-.071	-.034	.047	.121	.203	.282	.609
.664	-.159	-.041	-.034	-.026	-.025	.006	.047	.100	.161	.664
.719	.013	.009	.013	.022	.007	-.025	-.059	-.092	-.084	.719
.774	-.001	-.002	.001	.014	.017	-.006	-.118	-.229	-.264	.774
.830	-.007	-.006	-.003	.007	.010	-.004	-.108	-.274	-.355	.830
.871	-.007	-.010	-.010	-.004	-.002	-.012	-.081	-.200	-.356	.871
M = 0.980; q = 382 lb/sq ft										
.166	-.010	.005	.007	.024	.054	.072	.092	.110	.114	.166
.277										.277
.367	.013	.037	.035	.068	.085	.139	.203	.265	.314	.367
.387	-.056	-.002	.036	.065	.103	.192	.284	.371	.440	.387
.443	-.154	-.097	-.050	.011	.063	.186	.309	.427	.527	.443
.498	-.204	-.137	-.095	-.044	.007	.134	.257	.374	.475	.498
.553	-.203	-.145	-.110	-.052	-.003	.124	.236	.343	.438	.553
.609	-.306	-.248	-.205	-.146	-.065	.050	.148	.241	.326	.609
.664	-.316	-.257	-.207	-.138	-.066	.004	.074	.146	.211	.664
.719	-.126	-.116	-.090	-.078	-.083	-.100	-.080	-.025	-.027	.719
.774	-.016	-.002	.002	-.013	-.045	-.126	-.111	-.121	-.193	.774
.830	.008	.023	.018	.015	.001	-.088	-.168	-.219	-.301	.830
.871	.011	.024	.014	.013	.004	-.039	-.189	-.263	-.358	.871

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(d) Station D - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.166	.036	.050	.057	.072	.104	.107	.122	.137	.149	.166
.277										.277
.367	.069	.064	.055	.088	.116	.185	.250	.312	.361	.367
.387	.003	.051	.089	.120	.143	.242	.334	.418	.485	.387
.443	-.095	-.053	-.004	.051	.107	.235	.359	.472	.572	.443
.498	-.139	-.091	-.046	.004	.059	.182	.305	.419	.519	.498
.553	-.133	-.094	-.051	-.003	.052	.173	.287	.389	.482	.553
.609	-.232	-.192	-.146	-.091	-.027	.101	.200	.292	.374	.609
.664	-.242	-.199	-.154	-.090	-.022	.061	.131	.199	.265	.664
.719	-.114	-.091	-.085	-.052	-.046	-.050	-.043	-.005	.033	.719
.774	-.062	-.041	-.037	-.044	-.054	-.088	-.096	-.041	-.040	.774
.830	-.066	-.056	-.053	-.050	-.064	-.103	-.123	-.149	-.213	.830
.871	-.083	-.072	-.076	-.070	-.085	-.112	-.119	-.222	-.314	.871
M = 1.135; q = 425 lb/sq ft										
.166	.002	.020	.029	.038	.049	.071	.083	.084	.054	.166
.277										.277
.367	-.017	-.006	.000	-.013	-.011	-.021	-.045	-.024	.004	.367
.387	-.013	.000	.011	.038	.085	.156	.292	.426	.352	.387
.443	-.073	-.033	.006	.053	.094	.217	.360	.499	.531	.443
.498	-.106	-.061	-.023	.026	.078	.183	.315	.457	.592	.498
.553	-.089	-.053	-.013	.024	.071	.183	.308	.443	.564	.553
.609	-.185	-.151	-.108	-.063	-.011	.116	.242	.364	.470	.609
.664	-.193	-.153	-.108	-.066	-.008	.107	.199	.294	.375	.664
.719	-.098	-.082	-.067	-.049	-.028	.003	.023	.078	.157	.719
.774	-.047	-.047	-.047	-.044	-.040	-.042	-.045	.053	.055	.774
.830	-.039	-.046	-.051	-.055	-.052	-.062	-.073	-.068	-.103	.830
.871	-.044	-.051	-.059	-.062	-.064	-.078	-.065	-.105	-.149	.871
M = 1.200; q = 443 lb/sq ft										
.166	.017	.028	.039	.051	.055	.073	.092	.094	.095	.166
.277										.277
.367	-.011	-.000	.007	-.005	-.002	-.010	-.024	-.044	-.059	.367
.387	-.003	.008	.024	.035	.065	.104	.148	.318	.493	.387
.443	-.062	-.016	.027	.062	.106	.218	.346	.474	.604	.443
.498	-.089	-.041	.002	.048	.096	.200	.313	.436	.560	.498
.553	-.064	-.025	.011	.052	.100	.202	.299	.427	.541	.553
.609	-.146	-.120	-.080	-.040	.002	.103	.222	.357	.457	.609
.664	-.178	-.142	-.099	-.055	-.007	.104	.222	.307	.378	.664
.719	-.106	-.089	-.070	-.051	-.034	.021	.052	.089	.136	.719
.774	-.062	-.055	-.049	-.042	-.039	-.030	-.025	-.018	-.019	.774
.830	-.046	-.044	-.046	-.048	-.051	-.057	-.078	-.086	-.106	.830
.871	-.051	-.047	-.049	-.054	-.063	-.076	-.107	-.130	-.158	.871

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(e) Station E

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 310 lb/sq ft										
.055	.029	.044	.066	.091	.117	.178	.248	.331	.418	.055
.166	-.017	-.009	.003	.025	.046	.095	.152	.228	.304	.166
.277	-.025	-.024	-.015	.007	.026	.076	.137	.215	.300	.277
.367	-.076	-.053	-.033	-.001	.026	.098	.182	.279	.369	.367
.387	-.108	-.068	-.039	.007	.035	.119	.210	.309	.403	.387
.443	-.160	-.103	-.057	-.006	.037	.136	.235	.331	.419	.443
.498										.498
.553	-.160	-.114	-.074	-.027	.009	.094	.167	.239	.306	.553
.609	-.124	-.091	-.061	-.027	.001	.069	.120	.168	.216	.609
.664	-.065	-.049	-.033	-.008	.008	.049	.071	.086	.099	.664
.719	-.022	-.017	-.009	.005	.010	.032	.025	.001	-.044	.719
.774	-.007	-.008	-.007	.001	.003	.015	.002	-.037	-.105	.774
.830	-.001	-.005	-.007	-.002	-.004	-.000	-.009	-.033	-.089	.830
.871	.028	.022	.020	.025	.023	.028	.025	.008	-.027	.871
.954	.056	.050	.044	.045	.037	.031	.020	.026	.008	.954
M = 0.900; q = 354 lb/sq ft										
.055	.027	.049	.073	.091	.124	.191	.260	.344	.435	.055
.166	-.027	-.015	.003	.017	.045	.099	.158	.231	.319	.166
.277	-.032	-.023	-.018	-.003	.020	.074	.139	.222	.315	.277
.367	-.062	-.046	-.023	-.003	.028	.105	.191	.292	.398	.367
.387	-.096	-.062	-.029	.006	.041	.131	.226	.328	.434	.387
.443	-.181	-.119	-.063	-.017	.037	.145	.249	.352	.457	.443
.498										.498
.553	-.265	-.176	-.108	-.061	-.009	.091	.170	.255	.347	.553
.609	-.183	-.130	-.083	-.049	-.012	.062	.116	.182	.260	.609
.664	-.080	-.060	-.037	-.021	.006	.041	.053	.087	.144	.664
.719	-.025	-.016	-.005	.001	.016	.028	-.020	-.058	-.036	.719
.774	-.016	-.011	-.006	-.003	.007	.011	-.060	-.152	-.190	.774
.830	-.015	-.011	-.008	-.008	-.001	-.004	-.053	-.130	-.251	.830
.871	.015	.016	.017	.017	.022	.024	-.001	-.055	-.126	.871
.954	.049	.049	.048	.041	.041	.034	.022	.004	-.038	.954
M = 0.940; q = 367 lb/sq ft										
.055	.039	.055	.075	.104	.131	.195	.266	.358	.449	.055
.166	-.019	-.013	.001	.021	.045	.097	.160	.241	.330	.166
.277	-.027	-.021	-.023	-.002	.017	.073	.144	.234	.328	.277
.367	-.036	-.028	-.014	.008	.032	.111	.203	.308	.414	.367
.387	-.065	-.045	-.018	.022	.050	.142	.240	.348	.454	.387
.443	-.163	-.109	-.065	-.011	.038	.151	.265	.373	.477	.443
.498										.498
.553	-.257	-.214	-.153	-.085	-.028	.082	.181	.278	.370	.553
.609	-.308	-.248	-.124	-.069	-.034	.049	.124	.207	.288	.609
.664	-.188	-.048	-.039	-.026	-.021	.012	.055	.114	.178	.664
.719	.003	-.004	.002	.009	.004	-.016	-.035	-.034	.003	.719
.774	.000	-.006	-.004	.008	.011	-.009	-.103	-.167	-.164	.774
.830	-.005	-.010	-.009	-.001	.004	-.012	-.111	-.244	-.270	.830
.871	.019	.016	.019	.023	.028	.022	-.034	-.156	-.249	.871
.954	.058	.054	.050	.049	.045	.037	.017	-.003	-.068	.954
M = 0.980; q = 382 lb/sq ft										
.055	.050	.076	.089	.117	.142	.207	.283	.375	.472	.055
.166	-.016	-.003	.003	.033	.050	.104	.173	.261	.354	.166
.277	-.019	-.010	-.019	.004	.022	.080	.159	.257	.357	.277
.367	-.005	.008	.005	.030	.050	.126	.226	.338	.449	.367
.387	-.032	-.002	.009	.044	.073	.163	.267	.381	.489	.387
.443	-.134	-.081	-.042	.009	.055	.170	.290	.405	.511	.443
.498										.498
.553	-.237	-.183	-.145	-.091	-.042	.091	.204	.313	.409	.553
.609	-.297	-.241	-.198	-.140	-.067	.053	.150	.246	.331	.609
.664	-.316	-.258	-.208	-.140	-.067	.011	.082	.159	.226	.664
.719	-.152	-.135	-.100	-.080	-.078	-.077	-.036	.022	.060	.719
.774	-.015	-.002	-.000	-.014	-.046	-.113	-.082	-.077	-.091	.774
.830	.009	.024	.015	.006	-.008	-.097	-.151	-.180	-.216	.830
.871	.035	.048	.038	.041	.033	.002	-.127	-.179	-.222	.871
.954	.077	.085	.068	.068	.058	.044	-.036	-.251	-.316	.954

TABLE V.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 0.5 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Concluded

(e) Station E - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 400 lb/sq ft										
.055	.088	.108	.126	.155	.179	.246	.325	.410	.503	.055
.166	.032	.044	.052	.079	.099	.141	.204	.289	.386	.166
.277	-.001	.013	.009	.034	.048	.081	.193	.291	.394	.277
.367	.049	.042	.033	.060	.080	.172	.272	.381	.489	.367
.387	.028	.043	.053	.094	.115	.211	.318	.425	.530	.387
.443	-.072	-.042	-.002	.051	.101	.219	.340	.449	.556	.443
.498										.498
.553	-.167	-.133	-.089	-.038	.015	.138	.253	.360	.456	.553
.609	-.218	-.186	-.143	-.087	-.021	.102	.204	.295	.380	.609
.664	-.240	-.198	-.155	-.094	-.026	.065	.139	.211	.283	.664
.719	-.144	-.117	-.103	-.061	-.042	-.025	.009	.066	.127	.719
.774	-.060	-.039	-.036	-.044	-.052	-.072	-.061	-.001	.022	.774
.830	-.065	-.061	-.056	-.058	-.072	-.100	-.102	-.104	-.124	.830
.871	-.048	-.043	-.045	-.034	-.047	-.063	-.046	-.120	-.153	.871
.954	-.050	-.056	-.061	-.073	-.101	-.122	-.147	-.207	-.246	.954
M = 1.125; q = 425 lb/sq ft										
.055	.064	.082	.099	.119	.151	.214	.284	.366	.452	.055
.166	-.000	.014	.026	.043	.073	.116	.169	.241	.311	.166
.277	-.011	-.004	-.005	.005	.033	.067	.116	.178	.247	.277
.367	-.034	-.038	-.034	-.028	-.017	.014	.054	.108	.458	.367
.387	-.005	-.015	-.014	-.003	-.008	.010	.171	.386	.554	.387
.443	-.053	-.014	.009	.050	.088	.205	.337	.474	.612	.443
.498										.498
.553	-.119	-.086	-.051	-.011	.038	.149	.280	.415	.537	.553
.609	-.167	-.136	-.095	-.057	-.008	.116	.245	.367	.474	.609
.664	-.190	-.150	-.105	-.064	-.012	.107	.207	.305	.389	.664
.719	-.153	-.123	-.096	-.066	-.029	.035	.081	.157	.241	.719
.774	-.047	-.050	-.048	-.047	-.036	-.024	-.003	.112	.132	.774
.830	-.035	-.044	-.053	-.060	-.059	-.058	-.041	-.013	-.002	.830
.871	-.015	-.019	-.026	-.027	-.028	-.022	.006	-.021	-.020	.871
.954	.009	-.002	-.032	-.054	-.059	-.079	-.040	-.073	-.077	.954
M = 1.200; q = 443 lb/sq ft										
.055	.075	.095	.112	.135	.164	.228	.292	.380	.468	.055
.166	.012	.025	.036	.055	.068	.126	.182	.251	.327	.166
.277	.001	.009	.011	.020	.038	.080	.134	.190	.260	.277
.367	-.036	-.020	-.022	-.011	.002	.033	.074	.124	.182	.367
.387	-.017	-.013	-.010	-.001	-.012	.014	.048	.102	.376	.387
.443	-.035	-.002	.028	.082	.110	.206	.327	.451	.577	.443
.498										.498
.553	-.106	-.064	-.024	.016	.057	.160	.270	.400	.516	.553
.609	-.144	-.101	-.081	-.038	.004	.112	.223	.362	.462	.609
.664	-.172	-.128	-.084	-.041	.002	.103	.222	.317	.393	.664
.719	-.160	-.128	-.099	-.065	-.029	.057	.126	.179	.237	.719
.774	-.062	-.055	-.052	-.044	-.035	-.005	.020	.044	.083	.774
.830	-.036	-.035	-.035	-.038	-.040	-.038	-.038	-.014	.005	.830
.871	-.022	-.019	-.021	-.019	-.021	-.015	-.008	-.019	-.005	.871
.954	-.043	-.049	-.063	-.073	-.080	-.097	-.111	-.124	-.125	.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE

OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING

(a) Station A

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 675 lb/sq ft										
.055	.122	.091	.071	.046	.028	-.002	-.022	-.026	-.028	.055
.166	.047	.021	.010	-.006	-.017	-.027	-.034	-.031	-.040	.166
.277	.029	.005	-.006	-.020	-.028	-.034	-.027	-.011	-.016	.277
.367	.061	.023	.002	-.025	-.048	-.094	-.147	-.206	-.230	.367
.387	.066	.022	-.009	-.045	-.076	-.148	-.239	-.337	-.362	.387
.415	.068	.017	-.024	-.070	-.113	-.204	-.308	-.402	-.443	.415
.443	.049	-.006	-.050	-.102	-.150	-.252	-.366	-.397	-.465	.443
.498	-.002	-.057	-.102	-.154	-.205	-.319	-.425	-.469	-.464	.498
.553	-.031	-.079	-.120	-.166	-.208	-.302	-.338	-.428	-.399	.553
.581	-.044	-.085	-.120	-.159	-.196	-.270	-.325	-.410	-.410	.581
.609	-.047	-.082	-.110	-.143	-.171	-.226	-.292	-.396	-.413	.609
.636	-.033	-.066	-.087	-.112	-.132	-.166	-.230	-.349	-.397	.636
.664	-.021	-.045	-.059	-.076	-.089	-.107	-.165	-.287	-.384	.664
.692	-.011	-.027	-.034	-.048	-.053	-.061	-.113	-.221	-.354	.692
.719	-.004	-.016	-.020	-.028	-.030	-.031	-.073	-.172	-.325	.719
.774	-.007	-.012	-.011	-.015	-.012	-.010	-.035	-.106	-.235	.774
.830	-.004	-.008	-.004	-.005	-.001	.003	-.011	-.043	-.110	.830
.871	.010	.007	.011	.012	.018	.028	.019	.002	-.028	.871
.954	.026	.030	.039	.041	.046	.045	.038	.033	.046	.954
M = 0.900; q = 714 lb/sq ft										
.055	.127	.096	.072	.049	.029	.000	-.012	-.012	-.013	.055
.166	.043	.019	.003	-.011	-.022	-.029	-.027	-.018	-.025	.166
.277	.022	-.002	-.017	-.027	-.033	-.028	-.005	.026	.027	.277
.367	.069	.035	.011	-.008	-.024	-.051	-.070	-.110	-.153	.367
.387	.079	.037	.004	-.026	-.054	-.107	-.163	-.241	-.304	.387
.415	.077	.026	-.018	-.059	-.099	-.175	-.253	-.358	-.419	.415
.443	.052	-.005	-.052	-.098	-.144	-.226	-.319	-.419	-.411	.443
.498	-.014	-.074	-.129	-.184	-.235	-.323	-.421	-.454	-.487	.498
.553	-.059	-.117	-.176	-.244	-.309	-.410	-.490	-.443	-.481	.553
.581	-.072	-.124	-.179	-.254	-.332	-.447	-.513	-.496	-.474	.581
.609	-.075	-.120	-.162	-.217	-.338	-.479	-.518	-.540	-.469	.609
.636	-.059	-.095	-.124	-.155	-.200	-.481	-.462	-.519	-.435	.636
.664	-.036	-.061	-.078	-.092	-.097	-.231	-.146	-.474	-.417	.664
.692	-.017	-.031	-.040	-.048	-.047	-.031	-.105	-.382	-.435	.692
.719	-.006	-.017	-.021	-.024	-.021	-.003	-.094	-.180	-.334	.719
.774	-.009	-.016	-.015	-.015	-.009	.004	-.070	-.186	-.336	.774
.830	-.012	-.013	-.010	-.009	-.003	.005	-.031	-.087	-.210	.830
.871	.003	.002	.006	.009	.015	.028	.013	-.023	-.107	.871
.954	.026	.030	.038	.042	.045	.043	.040	.023	-.011	.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Continued

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.940; q = 732 lb/sq ft										
.055	.129	.103	.078	.060	.042	.014	-.004	.000	.004	.055
.166	.040	.020	.003	-.006	-.014	-.022	-.021	-.006	-.007	.166
.277	.017	-.003	-.020	-.023	-.026	-.017	.009	.048	.056	.277
.367	.078	.047	.025	.013	.002	-.015	-.031	-.061	-.102	.367
.387	.091	.052	.019	-.003	-.026	-.071	-.119	-.180	-.258	.387
.415	.086	.038	-.005	-.040	-.074	-.147	-.213	-.304	-.372	.415
.443	.056	.005	-.043	-.080	-.121	-.201	-.281	-.375	-.460	.443
.498	-.020	-.078	-.131	-.172	-.213	-.295	-.386	-.473	-.385	.498
.553	-.081	-.146	-.206	-.256	-.299	-.384	-.459	-.492	-.491	.553
.581	-.099	-.171	-.238	-.285	-.333	-.425	-.495	-.423	-.504	.581
.609	-.105	-.167	-.260	-.313	-.360	-.462	-.496	-.503	-.551	.609
.636	-.086	-.129	-.242	-.321	-.369	-.469	-.515	-.506	-.543	.636
.664	-.052	-.065	-.092	-.275	-.379	-.483	-.525	-.471	-.494	.664
.692	-.018	-.023	-.023	-.038	-.135	-.485	-.509	-.489	-.457	.692
.719	-.001	-.006	-.006	.007	.002	-.058	-.077	-.335	-.390	.719
.774	-.007	-.009	-.008	.007	.022	.030	-.040	-.249	-.304	.774
.830	-.014	-.012	-.010	.003	.018	.039	-.031	-.103	-.272	.830
.871	-.000	.003	.004	.015	.030	.055	.007	-.051	-.118	.871
.954	.029	.036	.040	.052	.061	.071	.047	.020	-.042	.954
M = 0.980; q = 776 lb/sq ft										
.055	.146	.116	.094	.073	.055	.027	.013			.055
.166	.050	.027	.012	-.000	-.006	-.013	-.008			.166
.277	.025	.000	-.012	-.020	-.022	-.007	.030			.277
.367	.120	.090	.071	.044	.035	.022	.013			.367
.387	.122	.085	.056	.030	.009	-.029	-.066			.387
.415	.115	.066	.029	-.007	-.039	-.105	-.164			.415
.443	.083	.034	-.008	-.048	-.085	-.160	-.232			.443
.498	.001	-.051	-.095	-.133	-.170	-.249	-.336			.498
.553	-.072	-.129	-.176	-.220	-.259	-.329	-.407			.553
.581	-.110	-.160	-.208	-.253	-.290	-.368	-.452			.581
.609	-.134	-.189	-.238	-.282	-.325	-.408	-.488			.609
.636	-.142	-.199	-.251	-.296	-.336	-.417	-.505			.636
.664	-.159	-.215	-.266	-.315	-.356	-.431	-.513			.664
.692	-.156	-.217	-.268	-.313	-.363	-.451	-.539			.692
.719	-.099	-.125	-.156	-.180	-.218	-.343	-.506			.719
.774	-.013	.003	.008	.003	-.021	-.128	-.167			.774
.830	-.007	.013	.026	.031	.029	-.011	-.034			.830
.871	.014	.026	.038	.046	.052	.050	.023			.871
.954	.050	.058	.070	.081	.090	.104	.084			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Continued

x/l	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/l
M = 1.050; q = 810 lb/sq ft										
.055	.179	.151	.129	.108	.089	.065	.062			.055
.166	.092	.075	.057	.047	.038	.028	.024			.166
.277	.044	.033	.013	.003	-.001	-.009	.075			.277
.367	.164	.133	.101	.089	.086	.099	.107			.367
.387	.171	.133	.096	.074	.058	.040	.013			.387
.415	.165	.116	.068	.034	.006	-.034	-.083			.415
.443	.135	.082	.033	-.006	-.039	-.091	-.155			.443
.498	.057	.001	-.045	-.083	-.117	-.176	-.255			.498
.553	-.015	-.076	-.126	-.164	-.199	-.255	-.324			.553
.581	-.047	-.105	-.156	-.196	-.230	-.290	-.365			.581
.609	-.074	-.132	-.185	-.225	-.258	-.327	-.403			.609
.636	-.086	-.140	-.195	-.237	-.271	-.340	-.418			.636
.664	-.104	-.153	-.214	-.256	-.288	-.356	-.429			.664
.692	-.112	-.156	-.219	-.264	-.300	-.379	-.452			.692
.719	-.087	-.107	-.152	-.173	-.197	-.310	-.449			.719
.774	-.057	-.048	-.037	-.047	-.057	-.105	-.130			.774
.830	-.068	-.057	-.052	-.053	-.062	-.081	-.082			.830
.871	-.072	-.055	-.054	-.052	-.054	-.049	-.028			.871
.954	-.125	-.103	-.067	-.045	-.036	-.021	-.045			.954
M = 1.130; q = 859 lb/sq ft										
.055	.153	.129	.105	.082	.059	.033	.008			.055
.166	.060	.045	.034	.019	.005	-.010	-.015			.166
.277	.030	.018	.005	-.007	-.012	-.024	-.028			.277
.367	-.012	-.022	-.027	-.033	-.030	-.023	-.008			.367
.387	.079	.061	.049	.036	.031	.012	.016			.387
.415	.156	.111	.076	.043	.013	-.034	-.057			.415
.443	.141	.095	.048	.009	-.026	-.089	-.120			.443
.498	.076	.034	-.007	-.047	-.083	-.150	-.199			.498
.553	.000	-.043	-.082	-.119	-.150	-.213	-.263			.553
.581	-.020	-.068	-.107	-.147	-.179	-.247	-.291			.581
.609	-.042	-.089	-.134	-.175	-.211	-.278	-.318			.609
.636	-.057	-.102	-.146	-.187	-.221	-.283	-.335			.636
.664	-.066	-.116	-.156	-.192	-.224	-.294	-.349			.664
.692	-.076	-.119	-.163	-.205	-.240	-.307	-.366			.692
.719	-.078	-.114	-.146	-.179	-.209	-.291	-.378			.719
.774	-.045	-.050	-.050	-.053	-.053	-.067	-.089			.774
.830	-.057	-.053	-.049	-.048	-.043	-.045	-.044			.830
.871	-.041	-.042	-.037	-.035	-.028	-.020	-.013			.871
.954	-.075	-.063	-.043	-.026	-.006	.008	.029			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(a) Station A - Concluded

x/l	$\alpha = -4^\circ$	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$	$\alpha = 8^\circ$	$\alpha = 12^\circ$	$\alpha = 16^\circ$	$\alpha = 20^\circ$	x/l
M = 1.200; q = 889 lb/sq ft										
.055	.161	.138	.109	.088	.072	.045	.017			.055
.166	.072	.057	.036	.028	.017	-.001	-.004			.166
.277	.034	.020	.003	.001	.003	-.014	-.012			.277
.367	-.015	-.022	-.040	-.041	-.032	-.036	-.042			.367
.387	.015	.011	.004	.014	.016	.008	-.001			.387
.415	.145	.113	.079	.051	.026	-.011	-.042			.415
.443	.146	.103	.053	.020	-.010	-.069	-.106			.443
.498	.103	.060	.012	-.026	-.065	-.127	-.174			.498
.553	.028	-.012	-.058	-.091	-.121	-.183	-.230			.553
.581	-.004	-.042	-.086	-.118	-.153	-.208	-.261			.581
.609	-.020	-.064	-.114	-.151	-.181	-.242	-.285			.609
.636	-.041	-.080	-.124	-.151	-.188	-.253	-.303			.636
.664	-.046	-.086	-.137	-.174	-.205	-.264	-.313			.664
.692	-.071	-.108	-.148	-.181	-.210	-.269	-.329			.692
.719	-.081	-.120	-.160	-.188	-.219	-.284	-.354			.719
.774	-.065	-.068	-.075	-.076	-.078	-.084	-.091			.774
.830	-.060	-.060	-.064	-.056	-.047	-.038	-.027			.830
.871	-.042	-.040	-.042	-.032	-.024	-.012	-.012			.871
.954	-.082	-.072	-.065	-.046	-.031	-.029	-.037			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(b) Station B

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 625 lb/sq ft										
.166	.026	.018	.011	-.002	-.014	-.043	-.079	-.102	-.121	.166
.277	.005	-.005	-.009	-.021	-.032	-.055	-.078	-.085	-.097	.277
.367	.090	.056	.033	-.003	-.038	-.120	-.211	-.274	-.298	.367
.387	.113	.062	.020	-.034	-.087	-.204	-.336	-.466	-.515	.387
.443	.056	-.005	-.058	-.119	-.179	-.306	-.464	-.499	-.666	.443
.498	-.009	-.063	-.113	-.172	-.228	-.365	-.512	-.553	-.616	.498
.553	-.059	-.107	-.148	-.197	-.245	-.358	-.424	-.536	-.625	.553
.609	-.056	-.091	-.121	-.155	-.185	-.252	-.339	-.473	-.632	.609
.664	-.031	-.050	-.062	-.078	-.089	-.113	-.198	-.368	-.702	.664
.719	-.003	-.012	-.012	-.017	-.018	-.017	-.077	-.205	-.413	.719
.774	-.023	-.027	-.023	-.025	-.022	-.018	-.059	-.124	-.219	.774
.830	-.013	-.016	-.011	-.012	-.009	-.004	-.019	-.039	-.094	.830
.871	-.000	-.001	.004	.004	.007	.009	.010	.003	-.033	.871
M = 0.900; q = 714 lb/sq ft										
.166	.027	.013	.005	-.006	-.018	-.044	-.070	-.087	-.105	.166
.277	-.001	-.012	-.019	-.028	-.036	-.051	-.054	-.052	-.058	.277
.367	.103	.071	.044	.017	-.012	-.068	-.118	-.157	-.206	.367
.387	.129	.079	.035	-.011	-.058	-.155	-.241	-.340	-.440	.387
.443	.058	-.002	-.058	-.114	-.168	-.275	-.402	-.535	-.561	.443
.498	-.022	-.083	-.140	-.200	-.258	-.364	-.475	-.461	-.610	.498
.553	-.086	-.134	-.205	-.271	-.341	-.455	-.568	-.537	-.648	.553
.609	-.084	-.129	-.172	-.229	-.354	-.503	-.533	-.609	-.648	.609
.664	-.047	-.067	-.081	-.092	-.095	-.200	-.224	-.526	-.664	.664
.719	-.004	-.010	-.010	-.013	-.008	.002	-.093	-.228	-.428	.719
.774	-.027	-.029	-.027	-.024	-.018	-.001	-.081	-.176	-.307	.774
.830	-.020	-.021	-.017	-.015	-.010	-.003	-.034	-.086	-.194	.830
.871	-.008	-.008	-.003	.000	.003	.006	-.000	-.035	-.122	.871
M = 0.940; q = 732 lb/sq ft										
.166	.024	.015	.004	-.002	-.011	-.037	-.065	-.075	-.088	.166
.277	-.005	-.014	-.023	-.026	-.030	-.041	-.041	-.029	-.031	.277
.367	.112	.084	.059	.038	.018	-.031	-.075	-.103	-.150	.367
.387	.142	.095	.050	.013	-.029	-.116	-.193	-.277	-.384	.387
.443	.064	.006	-.049	-.095	-.143	-.249	-.360	-.480	-.571	.443
.498	-.027	-.085	-.144	-.189	-.231	-.330	-.436	-.538	-.529	.498
.553	-.107	-.171	-.232	-.284	-.328	-.432	-.534	-.491	-.634	.553
.609	-.113	-.178	-.270	-.328	-.377	-.487	-.566	-.594	-.723	.609
.664	-.062	-.070	-.092	-.260	-.382	-.522	-.546	-.557	-.668	.664
.719	.001	.000	.002	.015	.005	-.084	-.139	-.306	-.339	.719
.774	-.023	-.023	-.020	-.003	.015	.020	-.068	-.243	-.262	.774
.830	-.021	-.019	-.016	-.005	.010	.025	-.039	-.126	-.248	.830
.871	-.011	-.007	-.004	.006	.016	.035	-.009	-.069	-.157	.871
M = 0.980; q = 776 lb/sq ft										
.166	.036	.023	.016	.005	-.005	-.028	-.051			.166
.277	.006	-.009	-.015	-.022	-.026	-.030	-.020			.277
.367	.141	.113	.090	.068	.049	.009	-.026			.367
.387	.172	.127	.086	.047	.008	-.069	-.139			.387
.443	.092	.037	-.011	-.060	-.107	-.203	-.310			.443
.498	-.005	-.057	-.105	-.149	-.190	-.280	-.385			.498
.553	-.094	-.150	-.201	-.243	-.287	-.376	-.486			.553
.609	-.144	-.195	-.250	-.296	-.339	-.431	-.536			.609
.664	-.179	-.240	-.288	-.335	-.382	-.471	-.576			.664
.719	-.092	-.107	-.128	-.142	-.176	-.250	-.331			.719
.774	-.031	-.011	-.003	-.008	-.036	-.144	-.218			.774
.830	-.012	.008	.020	.023	.021	-.035	-.071			.830
.871	.007	.019	.030	.035	.041	.031	-.009			.871

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(b) Station B - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 810 lb/sq ft										
.166	.079	.073	.061	.052	.041	.014	-.018			.166
.277	.026	.023	.009	.003	-.005	-.032	.026			.277
.367	.186	.156	.121	.102	.089	.075	.046			.367
.387	.222	.175	.125	.089	.057	.003	-.058			.387
.443	.142	.085	.027	-.018	-.062	-.131	-.227			.443
.498	.051	-.008	-.057	-.096	-.135	-.207	-.302			.498
.553	-.036	-.094	-.148	-.189	-.226	-.299	-.399			.553
.609	-.085	-.142	-.197	-.239	-.277	-.354	-.447			.609
.664	-.128	-.176	-.235	-.278	-.314	-.396	-.490			.664
.719	-.082	-.087	-.116	-.127	-.143	-.203	-.268			.719
.774	-.080	-.066	-.053	-.062	-.073	-.125	-.185			.774
.830	-.074	-.062	-.057	-.058	-.069	-.104	-.108			.830
.871	-.087	-.065	-.062	-.063	-.067	-.074	-.062			.871
M = 1.125; q = 859 lb/sq ft										
.166	.045	.041	.036	.025	.009	-.023	-.059			.166
.277	.013	.011	.007	-.003	-.012	-.041	-.075			.277
.367	-.010	-.001	-.002	-.010	-.019	-.033	-.023			.367
.387	.174	.136	.100	.064	.035	-.027	-.065			.387
.443	.143	.095	.045	-.001	-.043	-.127	-.196			.443
.498	.067	.020	-.026	-.069	-.107	-.186	-.251			.498
.553	-.019	-.064	-.107	-.146	-.180	-.258	-.336			.553
.609	-.054	-.099	-.142	-.188	-.227	-.309	-.376			.609
.664	-.094	-.143	-.185	-.223	-.255	-.330	-.421			.664
.719	-.063	-.081	-.097	-.117	-.130	-.156	-.181			.719
.774	-.064	-.061	-.059	-.062	-.064	-.093	-.114			.774
.830	-.062	-.055	-.050	-.050	-.049	-.074	-.069			.830
.871	-.048	-.047	-.043	-.044	-.039	-.046	-.045			.871
M = 1.200; q = 889 lb/sq ft										
.166	.058	.052	.039	.031	.019	-.015	-.045			.166
.277	.018	.012	.002	.001	-.002	-.032	-.052			.277
.367	-.020	-.012	-.020	-.022	-.024	-.045	-.051			.367
.387	.159	.126	.088	.064	.044	-.016	-.066			.387
.443	.139	.096	.044	.004	-.032	-.112	-.182			.443
.498	.090	.047	-.003	-.039	-.080	-.158	-.223			.498
.553	.011	-.032	-.082	-.117	-.152	-.228	-.293			.553
.609	-.030	-.068	-.113	-.147	-.186	-.259	-.338			.609
.664	-.082	-.118	-.158	-.194	-.227	-.297	-.375			.664
.719	-.072	-.087	-.108	-.119	-.132	-.149	-.170			.719
.774	-.090	-.086	-.096	-.092	-.095	-.126	-.150			.774
.830	-.066	-.062	-.065	-.064	-.061	-.074	-.077			.830
.871	-.050	-.046	-.051	-.046	-.042	-.046	-.046			.871

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(c) Station C

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 625 lb/sq ft										
.055										.055
.166	-.008	.000	.007	.001	-.008	-.044	-.112	-.201	-.307	.166
.277	-.020	-.014	-.007	-.013	-.022	-.058	-.122	-.200	-.288	.277
.353										.353
.367	.295	.291	.300	.290	.277	.236	.149	-.046	-.250	.367
.692	.038	.037	.042	.038	.042	.036	-.028	-.071	-.235	.692
.719	.005	.005	.009	.007	.010	.014	-.014	-.102	-.282	.719
.774	.001	-.001	.004	.003	.007	.012	-.011	-.074	-.193	.774
.830	-.009	-.010	-.006	-.008	-.006	-.003	-.020	-.045	-.104	.830
.871	-.013	-.014	-.009	-.012	-.010	-.010	-.022	-.020	-.055	.871
.954										.954
M = 0.900; q = 714 lb/sq ft										
.055										.055
.166	-.009	-.003	.001	-.003	-.012	-.046	-.102	-.185	-.290	.166
.277	-.023	-.020	-.017	-.019	-.027	-.056	-.102	-.168	-.248	.277
.353										.353
.367	.324	.310	.310	.305	.297	.285	.249	.102	-.087	.367
.692	.039	.040	.041	.040	.046	.007	-.101	-.139	-.226	.692
.719	.008	.007	.011	.012	.020	.027	-.047	-.175	-.304	.719
.774	-.002	-.004	.000	.003	.011	.022	-.059	-.178	-.333	.774
.830	-.014	-.015	-.012	-.011	-.007	-.004	-.048	-.114	-.251	.830
.871	-.019	-.020	-.017	-.016	-.013	-.014	-.034	-.064	-.157	.871
.954										.954
M = 0.940; q = 732 lb/sq ft										
.055										.055
.166	-.012	-.001	.001	.001	-.005	-.037	-.095	-.174	-.275	.166
.277	-.027	-.021	-.020	-.018	-.023	-.048	-.091	-.149	-.219	.277
.353										.353
.367	.341	.329	.327	.327	.323	.323	.285	.154	-.020	.367
.692	.037	.051	.049	.033	-.014	-.129	-.211	-.185	-.209	.692
.719	.022	.022	.024	.034	.021	-.060	-.107	-.190	-.259	.719
.774	.005	.004	.005	.021	.031	.023	-.072	-.243	-.323	.774
.830	-.014	-.013	-.011	-.002	.009	.011	-.073	-.213	-.337	.830
.871	-.021	-.019	-.018	-.011	-.003	-.003	-.053	-.126	-.244	.871
.954										.954
M = 0.980; q = 776 lb/sq ft										
.055										.055
.166	.003	.007	.012	.008	.002	-.029	-.084			.166
.277	-.015	-.015	-.012	-.014	-.018	-.039	-.072			.277
.353										.353
.367	.367	.351	.346	.340	.339	.341	.325			.367
.692	-.103	-.102	-.104	-.109	-.112	-.169	-.246			.692
.719	-.084	-.082	-.086	-.089	-.117	-.179	-.228			.719
.774	-.006	.011	.013	.002	-.030	-.134	-.188			.774
.830	.001	.017	.022	.018	.008	-.055	-.121			.830
.871	-.001	.008	.015	.013	.012	-.017	-.090			.871
.954										.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(c) Station C - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 810 lb/sq ft										
.055										.055
.166	.046	.056	.055	.055	.047	.013	-.042			.166
.277	.005	.017	.012	.012	.005	-.032	-.032			.277
.353										.353
.367	.418	.395	.372	.368	.376	.405	.405			.367
.692	-.061	-.056	-.069	-.066	-.068	-.102	-.163			.692
.719	-.080	-.069	-.076	-.074	-.081	-.121	-.160			.719
.774	-.057	-.039	-.036	-.046	-.059	-.103	-.143			.774
.830	-.074	-.060	-.057	-.060	-.075	-.116	-.149			.830
.871	-.106	-.084	-.083	-.087	-.097	-.123	-.136			.871
.954										.954
M = 1.125; q = 859 lb/sq ft										
.055										.055
.166	.012	.023	.031	.028	.019	-.024	-.087			.166
.277	-.002	.009	.015	.011	.002	-.039	-.118			.277
.353										.353
.367	.323	.318	.312	.304	.303	.296	.267			.367
.692	-.043	-.040	-.039	-.044	-.045	-.071	-.121			.692
.719	-.060	-.055	-.054	-.059	-.058	-.074	-.088			.719
.774	-.045	-.038	-.035	-.040	-.043	-.066	-.093			.774
.830	-.058	-.050	-.049	-.055	-.060	-.086	-.116			.830
.871	-.068	-.063	-.062	-.067	-.070	-.092	-.112			.871
.954										.954
M = 1.200; q = 889 lb/sq ft										
.055										.055
.166	.029	.035	.034	.037	.026	-.017	-.077			.166
.277	.005	.012	.007	.010	.002	-.033	-.107			.277
.353										.353
.367	.297	.298	.283	.287	.286	.281	.251			.367
.692	-.047	-.044	-.049	-.048	-.048	-.065	-.097			.692
.719	-.067	-.062	-.065	-.062	-.061	-.065	-.072			.719
.774	-.059	-.051	-.054	-.051	-.055	-.073	-.091			.774
.830	-.064	-.059	-.065	-.068	-.076	-.096	-.129			.830
.871	-.064	-.060	-.067	-.066	-.072	-.099	-.132			.871
.954										.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(d) Station D

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 625 lb/sq ft										
.166	.012	.022	.038	.044	.052	.070	.075	.077	.074	.166
.277										.277
.367	-.066	-.030	.005	.030	.054	.100	.142	.182	.221	.367
.387	-.138	-.081	-.026	.017	.059	.138	.210	.272	.331	.387
.443	-.187	-.117	-.055	-.003	.049	.159	.260	.349	.437	.443
.498	-.199	-.136	-.078	-.031	.018	.122	.213	.297	.381	.498
.553	-.173	-.124	-.075	-.037	.005	.093	.165	.232	.301	.553
.609	-.126	-.092	-.057	-.029	.003	.069	.118	.158	.207	.609
.664	-.055	-.038	-.018	-.005	.014	.054	.065	.068	.075	.664
.719	-.008	-.003	.007	.011	.019	.037	.019	-.035	-.119	.719
.774	-.005	-.004	.002	.003	.007	.016	-.005	-.063	-.176	.774
.830	.000	-.001	.004	.003	.004	.008	-.009	-.044	-.130	.830
.871	-.002	-.006	-.004	-.007	-.008	-.008	-.025	-.048	-.116	.871
M = 0.900; q = 714 lb/sq ft										
.166	.010	.020	.033	.041	.050	.067	.084	.090	.095	.166
.277										.277
.367	-.038	-.016	.011	.035	.059	.108	.163	.209	.254	.367
.387	-.119	-.071	-.021	.023	.067	.151	.238	.304	.370	.387
.443	-.198	-.131	-.068	-.012	.045	.160	.274	.371	.470	.443
.498	-.241	-.173	-.108	-.052	.003	.115	.221	.314	.412	.498
.553	-.270	-.186	-.118	-.067	-.016	.080	.168	.248	.334	.553
.609	-.177	-.129	-.086	-.050	-.014	.056	.115	.171	.244	.609
.664	-.064	-.049	-.028	-.013	.007	.038	.051	.069	.120	.664
.719	-.006	-.001	.007	.013	.023	.034	-.020	-.093	-.123	.719
.774	-.009	-.007	-.002	.001	.009	.017	-.057	-.172	-.281	.774
.830	-.006	-.006	-.003	-.001	.003	.007	-.042	-.127	-.297	.830
.871	-.008	-.012	-.011	-.011	-.011	-.012	-.042	-.104	-.224	.871
M = 0.940; q = 732 lb/sq ft										
.166	.008	.023	.033	.046	.058	.079	.090	.101	.112	.166
.277										.277
.367	-.018	.004	.024	.049	.074	.126	.178	.233	.279	.367
.387	-.099	-.053	-.010	.037	.083	.174	.255	.330	.396	.387
.443	-.191	-.122	-.067	-.008	.053	.177	.285	.392	.492	.443
.498	-.231	-.174	-.121	-.061	.002	.124	.229	.335	.436	.498
.553	-.281	-.228	-.166	-.095	-.030	.080	.174	.268	.359	.553
.609	-.327	-.257	-.123	-.073	-.032	.045	.119	.194	.274	.609
.664	-.155	-.037	-.030	-.022	-.015	.007	.045	.094	.155	.664
.719	.012	.013	.016	.023	.021	-.020	-.052	-.085	-.080	.719
.774	-.000	-.000	.003	.015	.023	.010	-.084	-.221	-.262	.774
.830	-.006	-.003	-.002	.006	.017	.013	-.074	-.259	-.344	.830
.871	-.011	-.009	-.012	-.006	-.002	-.006	-.066	-.186	-.347	.871
M = 0.980; q = 776 lb/sq ft										
.166	.023	.032	.043	.054	.065	.083	.102			.166
.277										.277
.367	.024	.037	.055	.072	.095	.144	.202			.367
.387	-.052	-.015	.027	.067	.109	.193	.282			.387
.443	-.145	-.095	-.034	.018	.073	.190	.309			.443
.498	-.193	-.138	-.088	-.040	.015	.133	.249			.498
.553	-.245	-.194	-.146	-.094	-.043	.083	.190			.553
.609	-.295	-.245	-.194	-.144	-.071	.044	.133			.609
.664	-.307	-.256	-.198	-.140	-.073	.000	.060			.664
.719	-.110	-.093	-.082	-.065	-.081	-.099	-.110			.719
.774	-.002	.012	.010	-.001	-.034	-.128	-.156			.774
.830	.017	.027	.028	.022	.009	-.059	-.123			.830
.871	.019	.021	.020	.014	.009	-.026	-.090			.871

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(d) Station D - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 810 lb/sq ft										
.166	.067	.081	.087	.099	.108	.124	.140			.166
.277										.277
.367	.075	.083	.082	.101	.124	.195	.258			.367
.387	.004	.038	.064	.105	.148	.248	.339			.387
.443	-.088	-.048	-.003	.058	.114	.243	.362			.443
.498	-.133	-.089	-.043	.007	.062	.184	.302			.498
.553	-.180	-.139	-.096	-.046	.011	.132	.244			.553
.609	-.227	-.187	-.144	-.092	-.029	.096	.191			.609
.664	-.238	-.196	-.152	-.091	-.030	.058	.125			.664
.719	-.110	-.086	-.075	-.054	-.047	-.045	-.045			.719
.774	-.059	-.038	-.034	-.047	-.056	-.088	-.101			.774
.830	-.063	-.052	-.050	-.051	-.063	-.099	-.125			.830
.871	-.086	-.070	-.073	-.075	-.085	-.106	-.115			.871
M = 1.125; q = 859 lb/sq ft										
.166	.027	.043	.057	.068	.080	.085	.099			.166
.277										.277
.367	-.011	-.005	.002	.006	.006	-.013	-.035			.367
.387	-.016	.010	.035	.058	.088	.152	.286			.387
.443	-.070	-.030	.015	.059	.111	.223	.359			.443
.498	-.108	-.063	-.015	.032	.085	.186	.313			.498
.553	-.141	-.098	-.054	-.012	.038	.138	.269			.553
.609	-.187	-.146	-.098	-.054	-.003	.110	.234			.609
.664	-.188	-.144	-.096	-.056	-.000	.105	.193			.664
.719	-.098	-.079	-.061	-.045	-.024	.005	.025			.719
.774	-.050	-.043	-.036	-.036	-.033	-.042	-.050			.774
.830	-.045	-.039	-.039	-.043	-.048	-.068	-.082			.830
.871	-.052	-.051	-.053	-.057	-.059	-.075	-.092			.871
M = 1.200; q = 889 lb/sq ft										
.166	.042	.056	.063	.076	.084	.093	.100			.166
.277										.277
.367	-.007	-.001	-.003	.006	.010	-.005	-.016			.367
.387	-.004	.021	.032	.052	.071	.103	.142			.387
.443	-.055	-.016	.018	.062	.114	.219	.350			.443
.498	-.081	-.038	.003	.050	.102	.205	.318			.498
.553	-.112	-.070	-.034	.011	.059	.166	.267			.553
.609	-.143	-.112	-.073	-.029	.015	.104	.218			.609
.664	-.173	-.132	-.095	-.049	-.001	.095	.213			.664
.719	-.102	-.082	-.068	-.047	-.025	.015	.058			.719
.774	-.063	-.055	-.055	-.045	-.036	-.032	-.031			.774
.830	-.042	-.041	-.049	-.048	-.052	-.062	-.072			.830
.871	-.047	-.043	-.049	-.052	-.057	-.079	-.101			.871

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Continued

(e) Station E

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 0.800; q = 625 lb/sq ft										
.055	.026	.040	.066	.089	.112	.177	.242	.323	.411	.055
.166	-.010	-.001	.014	.028	.046	.097	.154	.226	.307	.166
.277	-.022	-.017	-.003	.011	.027	.078	.138	.215	.301	.277
.367	-.075	-.053	-.025	-.000	.029	.102	.183	.272	.368	.367
.387	-.108	-.072	-.034	-.002	.034	.120	.207	.303	.399	.387
.443	-.157	-.102	-.049	-.005	.041	.143	.237	.328	.421	.443
.498										.498
.553	-.163	-.118	-.072	-.034	.005	.091	.163	.229	.300	.553
.609	-.122	-.089	-.055	-.027	.003	.070	.118	.162	.214	.609
.664	-.062	-.046	-.024	-.010	.010	.054	.073	.082	.101	.664
.719	-.020	-.014	-.003	.003	.012	.035	.029	-.005	-.040	.719
.774	-.008	-.007	.000	.001	.006	.018	.004	-.035	-.098	.774
.830	-.001	-.004	.000	-.001	.001	.006	-.006	-.031	-.080	.830
.871	.032	.028	.032	.030	.032	.037	.030	.012	-.015	.871
.954	.055	.050	.050	.044	.040	.035	.022	.027	.016	.954
M = 0.900; q = 714 lb/sq ft										
.055	.029	.045	.068	.091	.120	.180	.254	.335	.427	.055
.166	-.011	-.006	.008	.024	.044	.093	.160	.231	.319	.166
.277	-.023	-.022	-.010	.002	.020	.071	.143	.221	.314	.277
.367	-.055	-.042	-.021	.000	.029	.101	.195	.286	.392	.367
.387	-.092	-.064	-.032	.000	.039	.125	.225	.323	.429	.387
.443	-.169	-.116	-.062	-.015	.036	.144	.254	.349	.453	.443
.498										.498
.553	-.262	-.178	-.112	-.064	-.014	.080	.167	.245	.334	.553
.609	-.171	-.125	-.084	-.049	-.013	.057	.118	.177	.251	.609
.664	-.072	-.057	-.035	-.017	.004	.041	.061	.084	.139	.664
.719	-.018	-.013	-.004	.004	.016	.032	-.004	-.050	-.038	.719
.774	-.011	-.010	-.005	-.001	.007	.017	-.042	-.135	-.186	.774
.830	-.008	-.008	-.006	-.006	-.002	.001	-.041	-.115	-.245	.830
.871	.025	.023	.025	.025	.030	.033	.011	-.042	-.115	.871
.954	.053	.051	.050	.047	.044	.036	.026	.010	-.031	.954
M = 0.940; q = 752 lb/sq ft										
.055	.032	.052	.073	.101	.131	.196	.261	.348	.440	.055
.166	-.013	-.003	.008	.026	.050	.103	.162	.242	.330	.166
.277	-.026	-.021	-.014	.001	.023	.078	.146	.233	.329	.277
.367	-.036	-.027	-.012	.012	.041	.118	.203	.307	.411	.367
.387	-.073	-.048	-.020	.014	.053	.144	.239	.344	.450	.387
.443	-.166	-.108	-.061	-.008	.047	.161	.265	.372	.476	.443
.498										.498
.553	-.270	-.218	-.162	-.092	-.029	.081	.173	.267	.360	.553
.609	-.318	-.250	-.122	-.071	-.031	.050	.122	.200	.279	.609
.664	-.183	-.042	-.036	-.024	-.014	.015	.057	.110	.174	.664
.719	.004	.002	.004	.013	.013	-.006	-.027	-.033	.002	.719
.774	-.002	-.003	-.001	.010	.018	.008	-.071	-.164	-.160	.774
.830	-.007	-.006	-.005	.001	.009	.005	-.076	-.236	-.262	.830
.871	.025	.025	.025	.031	.040	.041	-.011	-.135	-.231	.871
.954	.054	.056	.054	.054	.052	.045	.024	.001	-.042	.954
M = 0.980; q = 776 lb/sq ft										
.055	.052	.067	.090	.114	.143	.203	.277			.055
.166	-.001	.008	.020	.035	.056	.107	.174			.166
.277	-.011	-.013	-.005	.006	.027	.082	.158			.277
.367	.002	.005	.016	.033	.058	.130	.224			.367
.387	-.028	-.010	.015	.043	.077	.162	.263			.387
.443	-.122	-.082	-.029	.016	.066	.175	.290			.443
.498										.498
.553	-.233	-.190	-.143	-.093	-.040	.083	.190			.553
.609	-.287	-.241	-.192	-.140	-.071	.047	.137			.609
.664	-.305	-.254	-.199	-.142	-.072	.008	.071			.664
.719	-.135	-.111	-.093	-.067	-.076	-.079	-.062			.719
.774	.003	.014	.009	-.006	-.034	-.115	-.120			.774
.830	.021	.027	.022	.014	.001	-.068	-.131			.830
.871	.049	.051	.053	.049	.046	.022	-.023			.871
.954	.084	.080	.077	.070	.067	.054	.011			.954

TABLE VI.- PRESSURE COEFFICIENTS AT STAGNATION PRESSURE OF 1.0 ATMOSPHERE FOR BODY IN PRESENCE OF WING - Concluded

(e) Station E - Concluded

x/l	α = -4°	α = -2°	α = 0°	α = 2°	α = 4°	α = 8°	α = 12°	α = 16°	α = 20°	x/l
M = 1.030; q = 810 lb/sq ft										
.055	.087	.102	.122	.150	.174	.240	.321			.055
.166	.044	.055	.064	.081	.100	.146	.207			.166
.277	.005	.017	.018	.035	.051	.089	.194			.277
.367	.053	.050	.041	.057	.084	.176	.274			.367
.387	.027	.043	.054	.081	.118	.215	.318			.387
.443	-.067	-.035	.002	.056	.108	.227	.343			.443
.498										.498
.553	-.169	-.133	-.094	-.044	.012	.132	.244			.553
.609	-.218	-.182	-.142	-.090	-.026	.096	.194			.609
.664	-.235	-.194	-.153	-.094	-.030	.063	.134			.664
.719	-.143	-.111	-.098	-.058	-.046	-.024	.006			.719
.774	-.057	-.036	-.035	-.047	-.053	-.072	-.066			.774
.830	-.062	-.056	-.058	-.058	-.069	-.099	-.106			.830
.871	-.046	-.035	-.036	-.033	-.040	-.052	-.049			.871
.954	-.052	-.054	-.060	-.076	-.098	-.107	-.135			.954
M = 1.125; q = 859 lb/sq ft										
.055	.060	.079	.101	.124	.153	.210	.281			.055
.166	.006	.020	.037	.052	.075	.115	.178			.166
.277	-.003	.002	.014	.027	.045	.079	.127			.277
.367	-.041	-.038	-.031	-.023	-.008	.015	.053			.367
.387	-.007	-.006	-.005	-.005	-.002	.009	.094			.387
.443	-.045	-.026	.012	.053	.104	.213	.341			.443
.498										.498
.553	-.129	-.090	-.047	-.007	.040	.141	.272			.553
.609	-.172	-.132	-.088	-.047	.004	.111	.235			.609
.664	-.191	-.148	-.101	-.057	-.002	.105	.200			.664
.719	-.156	-.124	-.090	-.061	-.023	.039	.082			.719
.774	-.052	-.046	-.042	-.038	-.032	-.028	-.011			.774
.830	-.041	-.041	-.045	-.048	-.050	-.062	-.055			.830
.871	-.016	-.019	-.020	-.022	-.020	-.024	-.016			.871
.954	.001	-.020	-.036	-.052	-.062	-.080	-.085			.954
M = 1.200; q = 889 lb/sq ft										
.055	.074	.093	.109	.135	.164	.225	.296			.055
.166	.025	.037	.045	.065	.085	.129	.187			.166
.277	.007	.015	.020	.032	.049	.090	.140			.277
.367	-.033	-.026	-.027	-.011	.008	.034	.078			.367
.387	-.017	-.014	-.020	-.015	-.007	.016	.060			.387
.443	-.029	.001	.028	.068	.115	.214	.338			.443
.498										.498
.553	-.105	-.065	-.030	.014	.063	.157	.270			.553
.609	-.141	-.115	-.082	-.036	.017	.114	.220			.609
.664	-.170	-.127	-.085	-.032	.017	.105	.213			.664
.719	-.158	-.125	-.094	-.059	-.022	.051	.129			.719
.774	-.063	-.058	-.057	-.047	-.036	-.013	.015			.774
.830	-.035	-.037	-.042	-.039	-.041	-.036	-.039			.830
.871	-.016	-.015	-.020	-.016	-.016	-.019	-.007			.871
.954	-.045	-.052	-.066	-.071	-.080	-.097	-.108			.954

TABLE VII.- WING SECTION DATA

α, deg	$\frac{y}{b/2}$ = 0.12				$\frac{y}{b/2}$ = 0.25						$\frac{y}{b/2}$ = 0.40					
	c_n		c_m		$\Delta\alpha$, deg		c_n		c_m		$\Delta\alpha$, deg		c_n		c_m	
	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm
M = 0.800																
-4	-0.1962	-0.1993	0.0045	0.0059	0.02	0.04	-0.2207	-0.2199	-0.0113	-0.0083	0.06	0.11	-0.2610	-0.2586	-0.0196	-0.0179
-2	-.0847	-.0829	-.0053	-.0066	.01	.02	-.0841	-.0791	-.0156	-.0139	.03	.09	-.0982	-.0915	-.0237	-.0253
0	.0322	.0278	-.0202	-.0184	.00	-.01	.0479	.0429	-.0233	-.0219	-.01	-.01	.0575	.0544	-.0290	-.0280
2	.1384	.1357	-.0314	-.0297	-.01	-.03	.1694	.1660	-.0297	-.0286	-.04	-.07	.1881	.1864	-.0337	-.0334
4	.2490	.2443	-.0433	-.0415	-.03	-.05	.2947	.2880	-.0357	-.0349	-.06	-.12	.3287	.3202	-.0381	-.0382
8	.4785	.4730	-.0676	-.0654	-.04	-.09	.5604	.5588	-.0453	-.0432	-.11	-.22	.6262	.6256	-.0365	-.0319
12	.6836	.6851	-.0876	-.0866	-.05	-.10	.8018	.7956	-.0610	-.0636	-.13	-.25	1.0241	1.0257	-.1389	-.1295
16	.9167	.9155	-.1232	-.1191	-.06	-.11	1.1121	1.1159	-.1512	-.1402	-.14	-.27	1.0497	1.0416	-.1856	-.1818
20	1.1266	1.1296	-.1581	-.1547	-.06	-.12	1.1555	1.1659	-.2016	-.1942	-.16	-.31	1.0286	1.0186	-.1900	-.1871
M = 0.900																
-4	-0.2135	-0.2163	0.0120	0.0127	0.03	0.05	-0.2357	-0.2372	-0.0020	-0.0013	0.07	0.14	-0.2776	-0.2776	-0.0104	-0.0142
-2	-.0872	-.0894	-.0060	-.0059	.01	.02	-.0832	-.0866	-.0153	-.0142	.03	.06	-.0968	-.1001	-.0257	-.0269
0	.0361	.0311	-.0240	-.0222	-.01	-.01	.0559	.0485	-.0272	-.0256	-.01	-.02	.0658	.0617	-.0330	-.0317
2	.1552	.1480	-.0412	-.0381	-.02	-.04	.1898	.1808	-.0392	-.0366	-.05	-.09	.2146	.2062	-.0430	-.0404
4	.2800	.2712	-.0615	-.0570	-.03	-.06	.3368	.3276	-.0543	-.0507	-.09	-.16	.3799	.3626	-.0536	-.0491
8	.5291	.5220	-.1052	-.1028	-.06	-.12	.6246	.6168	-.0866	-.0866	-.15	-.30	.7014	.6932	-.0730	-.0662
12	.7292	.7148	-.1203	-.1193	-.06	-.12	.8497	.8443	-.1005	-.0927	-.16	-.31	1.0316	1.0429	-.1692	-.1546
16	.9286	.9329	-.1547	-.1567	-.07	-.14	1.1113	1.1268	-.1779	-.1642	-.18	-.34	1.0377	1.0453	-.1976	-.1948
20	1.1229	1.1407	-.1867	-.1812	-.08	-.15	1.1696	1.1937	-.2177	-.2145	-.20	-.39	1.0879	1.0868	-.2114	-.2084
M = 0.940																
-4	-0.2447	-0.2503	0.0353	0.0374	0.03	0.06	-0.2602	-0.2612	0.0129	0.0129	0.09	0.16	-0.3101	-0.3051	0.0040	-0.0041
-2	-.0874	-.0908	-.0059	-.0063	.01	.02	-.0973	-.0956	-.0115	-.0098	.04	.06	-.1107	-.1079	-.0240	-.0258
0	.0482	.0416	-.0339	-.0311	-.01	-.01	.0670	.0578	-.0362	-.0336	-.02	-.03	.0765	.0757	-.0412	-.0407
2	.1815	.1799	-.0641	-.0631	-.02	-.04	.2146	.2087	-.0583	-.0553	-.06	-.12	.2464	.2451	-.0610	-.0599
4	.3051	.3045	-.0883	-.0878	-.04	-.08	.3586	.3515	-.0797	-.0754	-.11	-.21	.4092	.3998	-.0807	-.0764
8	.5350	.5482	-.1279	-.1301	-.06	-.13	.6210	.6337	-.1075	-.1092	-.17	-.36	.7026	.7274	-.0999	-.1033
12	.7460	.7356	-.1449	-.1488	-.07	-.14	.8492	.8510	-.1198	-.1192	-.18	-.36	1.0624	1.0454	-.1617	-.1551
16	.9330	.9361	-.1674	-.1691	-.08	-.16	1.1428	1.1425	-.1780	-.1641	-.20	-.39	1.1020	1.1011	-.2159	-.2140
20	1.1483	1.1443	-.2069	-.2006	-.09	-.17	1.2711	1.2741	-.2357	-.2294	-.23	-.44	1.1893	1.1796	-.2409	-.2336
M = 0.980																
-4	-0.2219	-0.2192	0.0279	0.0266	0.03	0.06	-0.2500	-0.2441	0.0191	0.0191	0.09	0.16	-0.2928	-0.2698	0.0092	-0.0070
-2	-.0904	-.0884	.0019	.0014	.01	.03	-.0976	-.0987	-.0063	-.0035	.03	.08	-.1063	-.1091	-.0210	-.0214
0	.0254	.0244	-.0239	-.0235	-.01	-.01	.0462	.0421	-.0306	-.0283	-.02	-.02	.0605	.0615	-.0456	-.0453
2	.1440	.1409	-.0491	-.0470	-.03	-.05	.1848	.1770	-.0441	-.0408	-.08	-.15	.2254	.2185	-.0726	-.0687
4	.2692	.2629	-.0774	-.0740	-.05	-.09	.3327	.3193	-.0796	-.0747	-.13	-.23	.3942	.3780	-.0975	-.0904
8	.5087	.4975	-.1213	-.1164	-.08	-.15	.6074	.5898	-.1117	-.1083	-.22	-.40	.7029	.6824	-.1190	-.1161
12	.7436	.7302	-.1617	-.1584	-.09	-.20	.8664	.8619	-.1421	-.1425	-.25	-.52	1.0134	1.0141	-.1514	-.1554
16	.9508		-.1891		-.09		1.1114		-.1719		-.24		1.2890		-.2497	
20	1.1133		-.1912		-.10		1.3541		-.2313		-.27		1.3783		-.2665	
M = 1.030																
-4	-0.2107	-0.2087	0.0243	0.0237	0.03	0.06	-0.2395	-0.2328	0.0160	0.0158	0.09	0.17	-0.2801	-0.2682	0.0077	-0.0002
-2	-.1022	-.0966	.0018	.0016	.01	.03	-.1041	-.0964	-.0057	-.0051	.04	.08	-.1095	-.1044	-.0205	-.0205
0	.0237	.0197	-.0184	-.0220	-.01	-.01	.0433	.0393	-.0302	-.0284	-.02	-.03	.0606	.0556	-.0463	-.0444
2	.1389	.1342	-.0474	-.0451	-.03	-.05	.1807	.1712	-.0437	-.0401	-.08	-.14	.2204	.2096	-.0726	-.0680
4	.2555	.2468	-.0729	-.0686	-.04	-.09	.3160	.3035	-.0758	-.0711	-.12	-.22	.3794	.3592	-.0998	-.0874
8	.4842	.4751	-.1165	-.1117	-.08	-.15	.5775	.5680	-.1092	-.1066	-.22	-.41	.6740	.6555	-.1196	-.1167
12	.7132	.6964	-.1569	-.1528	-.10	-.20	.8363	.8247	-.1491	-.1387	-.28	-.54	.9656	.9720	-.1585	-.1539
16	.9409		-.1948		-.11		1.0789		-.1784		-.28		1.3623		-.2741	
20	1.1434		-.2254		-.11		1.2837		-.1904		-.29		1.4629		-.2963	
M = 1.125																
-4	-0.1921	-0.1910	0.0213	0.0207	0.03	0.06	-0.2092	-0.2037	0.0127	0.0118	0.09	0.17	-0.2492	-0.2381	0.0064	-0.0010
-2	-.0903	-.0845	-.0001	-.0004	.02	.03	-.0952	-.0910	-.0071	-.0099	.05	.08	-.1019	-.0929	-.0191	-.0245
0	.0177	.0191	-.0219	-.0214	.00	-.01	.0344	.0416	-.0270	-.0285	-.01	-.02	.0500	.0554	-.0441	-.0457
2	.1253	.1193	-.0430	-.0412	-.03	-.05	.1633	.1583	-.0484	-.0473	-.07	-.12	.1969	.1904	-.0686	-.0668
4	.2277	.2242	-.0657	-.0615	-.04	-.08	.2893	.2828	-.0683	-.0669	-.12	-.22	.3442	.3381	-.0899	-.0877
8	.4491	.4421	-.1077	-.1052	-.08	-.15	.5366	.5364	-.1055	-.1048	-.22	-.41	.6391	.6355	-.1255	-.1234
12	.6650	.6588	-.1488	-.1457	-.11	-.20	.7878	.7848	-.1414	-.1363	-.29	-.55	.9271	.9186	-.1523	-.1558
16	.8733		-.1862		-.11		1.0302		-.1733		-.27		1.2400		-.2486	
20	1.0822		-.2172		-.11		1.2103		-.1899		-.29		1.3651		-.2727	
M = 1.200																
-4	-0.1744	-0.1780	0.0198	0.0182	0.03	0.07	-0.1888	-0.1882	0.0100	0.0098	0.08	0.18	-0.2179	-0.2255	0.0010	-0.0038
-2	-.0945	-.0814	.0024	.0006	.02	.03	-.0999	-.0822	-.0065	-.0099	.05	.08	-.1154	-.0907	-.0169	-.0215
0	.0023	.0133	-.0196	-.0159	.00	-.01	.0145	.0355	-.0248	-.0281	.00	-.02	.0210	.0510	-.0369	-.0438
2	.1018	.1064	-.0359	-.0343	-.02	-.04	.1381	.1514	-.0447	-.0470	-.05	-.11	.1649	.1776	-.0594	-.0614
4	.2020	.2081	-.0536	-.0559	-.04	-.08	.2596	.2719	-.0648	-.0669	-.11	-.21	.3085	.3168	-.0805	-.0811
8	.4049	.4104	-.0951	-.0940	-.07	-.15	.4957	.5086	-.1020	-.1024	-.20	-.40	.5825	.5948	-.1189	-.1180
12	.6128	.6200	-.1386	-.1374	-.10	-.20	.7087	.7326	-.1376	-.1359	-.27	-.53	.8354	.8599	-.1581	-.1550
16	.8279		-.1817		-.12		.9621		-.1894		-.31		1.1147		-.2325	
20	1.0358		-.2127		-.13		1.1909		-.2293		-.33		1.3214		-.2935	

TABLE VII.- WING SECTION DATA - Concluded

α, deg	$\frac{y}{b/2}$ = 0.60						$\frac{y}{b/2}$ = 0.80						$\frac{y}{b/2}$ = 0.95					
	$\Delta\alpha$, deg		c_n		c_m		$\Delta\alpha$, deg		c_n		c_m		$\Delta\alpha$, deg		c_n		c_m	
	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm	0.5 atm	1.0 atm
M = 0.800																		
-4	0.15	0.28	-0.3680	-0.3572	-0.0064	-0.0078	0.27	0.46	-0.4891	-0.4499	0.0822	0.0532	0.29	0.49	-0.1676	-0.1774	-0.0063	-0.0036
-2	.08	.15	-.1361	-.1286	-.0278	-.0250	.14	.25	-.2414	-.2260	.0024	-.0351	.21	.39	-.3096	-.3480	.0457	.0591
0	-.03	-.04	.0624	.0552	-.0338	-.0320	-.06	-.06	-.0012	-.0141	-.0589	-.0267	.12	-.13	-.0860	-.1092	-.0463	-.0234
2	-.11	-.21	.3234	.2148	-.0375	-.0366	-.19	-.36	.1837	.1718	-.0426	-.0421	-.25	-.49	.1160	.1085	-.0390	-.0413
4	-.17	-.34	.3859	.3693	-.0388	-.0385	-.28	-.56	.3700	.3510	-.0390	-.0406	-.34	-.69	.2615	.2491	-.0340	-.0368
8	-.30	-.58	.8576	.8557	-.1178	-.1089	-.48	-.92	.5465	.5367	-.0846	-.0798	-.56	-1.08	.3104	.3128	-.0405	-.0403
12	-.34	-.67	.8032	.8314	-.1391	-.1407	-.53	-1.06	.5354	.5362	-.0849	-.0852	-.62	-1.24	.3257	.3262	-.0478	-.0485
16	-.37	-.68	.8013	.7885	-.1400	-.1364	-.59	-1.16	.5917	.5869	-.0970	-.0944	-.70	-1.38	.3827	.3829	-.0567	-.0581
20	-.42	-.83	.8525	.8378	-.1541	-.1510	-.68	-1.34	.6867	.6743	-.1178	-.1151	-.81	-1.61	.4570	.4521	-.0675	-.0699
M = 0.900																		
-4	0.20	0.36	-0.3863	-0.3839	-0.0104	-0.0112	0.35	0.60	-0.5381	-0.4919	0.1002	-0.0761	0.38	0.64	-0.1841	-0.1753	-0.0052	-0.0090
-2	.08	.17	-.1303	-.1417	-.0295	-.0277	.16	.27	-.2444	-.2455	.0034	-.0310	.22	.39	-.2865	-.3186	.0342	.0467
0	-.04	-.07	.0706	.0570	-.0371	-.0357	-.09	-.16	.0038	-.0025	-.0430	-.0420	-.17	-.34	-.0734	-.0892	-.0550	-.0580
2	-.15	-.26	.2539	.2347	-.0428	-.0403	-.25	-.46	.2182	.1899	-.0493	-.0461	-.34	-.63	.1491	.1274	-.0439	-.0453
4	-.23	-.44	.4338	.3668	-.0389	-.0365	-.37	-.71	.4342	.4049	-.0573	-.0394	-.43	-.86	.3427	.2975	-.0257	-.0398
8	-.39	-.77	.8977	.8465	-.1206	-.0691	-.64	-1.26	.6487	.7227	-.1093	-.1094	-.76	-1.52	.3529	.3926	-.0519	-.0586
12	-.43	-.84	.8236	.8189	-.1529	-.1461	-.68	-1.35	.5947	.5845	-.1042	-.1029	-.80	-1.61	.3730	.3741	-.0551	-.0590
16	-.48	-.93	.8486	.8354	-.1582	-.1534	-.77	-1.50	.6681	.6537	-.1173	-.1154	-.91	-1.79	.4205	.4246	-.0579	-.0641
20	-.59	-1.06	.9179	.9009	-.1773	-.1724	-.89	-1.72	.7657	.7472	-.1394	-.1363	-1.06	-2.07	.5017	.5016	-.0739	-.0777
M = 0.940																		
-4	0.26	0.43	-0.4202	-0.4206	0.0138	0.0110	0.48	0.70	-0.6062	-0.5139	0.1191	0.0518	0.55	0.77	-0.2752	-0.2561	0.0142	0.0063
-2	.10	.16	-.1444	-.1910	-.0304	-.0293	.19	.24	-.2845	-.2438	.0156	-.0183	.25	.34	-.2814	-.2452	.0320	.0328
0	-.02	-.09	.0786	.0691	-.0401	-.0393	-.11	-.19	.0158	.0043	-.0455	-.0455	-.20	-.38	-.0641	-.0872	-.0574	-.0585
2	-.18	-.33	.2957	.2782	-.0555	-.0518	-.31	-.56	.2609	.2314	-.0475	-.0449	-.39	-.75	.1762	.1546	-.0418	-.0470
4	-.32	-.59	.5150	.4828	-.0841	-.0743	-.52	-.96	.5026	.4721	-.0519	-.0516	-.59	-1.11	.4375	.3806	-.0235	-.0315
8	-.45	-.98	.8724	.8807	-.0992	-.0967	-.74	-1.55	.7546	.9916	-.1304	-.1236	-.88	-1.82	.4047	.4148	-.0559	-.0589
12	-.49	-.97	.8687	.9009	-.1665	-.1629	-.80	-1.58	.6786	.6556	-.1247	-.1188	-.96	-1.90	.4367	.4270	-.0662	-.0680
16	-.56	-1.09	.9319	.9073	-.1816	-.1747	-.92	-1.79	.7649	.7527	-.1423	-.1369	-1.10	-2.15	.5047	.4899	-.0783	-.0772
20	-.63	-1.23	1.0194	.9801	-.2018	-.1916	-1.04	-2.02	.8618	.8184	-.1644	-.1559	-1.26	-2.45	.5898	.5692	-.0927	-.0929
M = 0.980																		
-4	0.26	0.46	-0.3805	-0.3928	0.0280	0.0294	0.42	0.68	-0.5513	-0.4850	0.0399	-0.0019	0.50	0.76	-0.3562	-0.3288	0.0349	0.0248
-2	.09	.21	-.1446	-.1724	-.0245	-.0187	.14	.29	-.2710	-.2868	-.0314	-.0332	.19	.37	-.2734	-.2710	.0376	.0317
0	-.07	-.09	.0745	.0918	-.0592	-.0550	-.16	-.20	.0293	-.0167	-.0959	-.0768	-.23	-.30	-.1410	-.1814	-.0240	-.0163
2	-.28	-.50	.2853	.2896	-.0975	-.0808	-.56	-1.04	.3160	.2998	-.1295	-.1361	-.83	-1.49	.3114	.2426	-.1365	-.1053
4	-.40	-.74	.4922	.4488	-.1161	-.1042	-.75	-1.42	.5477	.4933	-.1471	-.1428	-1.00	-1.96	.4612	.4226	-.1142	-.1276
8	-.68	-1.23	.9169	.8386	-.1543	-.1359	-1.25	-2.22	1.0777	.9446	-.2273	-.1778	-1.59	-2.87	.8378	.7834	-.1434	-.1495
12	-.66	-1.54	1.1017	1.2092	-.2063	-.2122	-1.12	-2.64	.9562	1.1206	-.1789	-.1882	-1.37	-5.26	.6073	.9041	-.1077	-.1334
16	-.68		1.0404		-.2066		-1.13		.8799		-.1815		-1.58		.6123		-.0989	
20	-.76		1.1151		-.2344		-1.28		.9752		-.2029		-1.57		.7152		-.1230	
M = 1.030																		
-4	0.26	0.49	-0.3573	-0.3589	0.0239	0.0248	0.42	0.73	-0.5130	-0.4910	0.0273	0.0125	0.48	0.82	-0.3926	-0.3527	0.0192	0.0093
-2	.11	.21	-.1417	-.1543	-.0259	-.0250	.16	.29	-.2649	-.2610	-.0530	-.0534	.21	.38	-.3870	-.3668	.0380	.0322
0	-.09	-.14	.0759	.0584	-.0666	-.0624	-.23	-.40	.0139	-.0141	-.0954	-.0920	-.47	-.86	-.0736	-.1118	-.1322	-.1308
2	-.26	-.47	.2762	.2512	-.0980	-.0902	-.52	-.95	.2751	.2361	-.1209	-.1147	-.76	-1.41	.2043	.1685	-.1241	-.1142
4	-.39	-.71	.4719	.4362	-.1173	-.1091	-.72	-1.34	.4874	.4348	-.1345	-.1287	-.96	-1.81	.4034	.3429	-.1105	-.1100
8	-.68	-1.25	.8572	.8130	-.1487	-.1371	-1.24	-2.25	1.0022	.8965	-.2148	-.1743	-1.60	-2.94	.8154	.7531	-.1544	-.1516
12	-.80	-1.62	1.2174	1.1451	-.2383	-.2025	-1.48	-2.86	1.1971	1.0971	-.2127	-.1980	-1.88	-3.68	.9688	.9816	-.1727	-.1786
16	-.79		1.1797		-.2355		-1.32		.9640		-.1920		-1.63		.7156		-.1292	
20	-.83		1.2068		-.2465		-1.40		.9694		-.2048		-1.75		.7841		-.1463	
M = 1.120																		
-4	0.26	0.53	-0.3017	-0.3027	0.0151	0.0165	0.46	0.89	-0.4461	-0.4571	0.0223	0.0034	0.58	1.04	-0.5058	-0.5389	0.0501	0.0285
-2	.13	.21	-.1370	-.1541	-.0224	-.0252	.23	.28	-.2694	-.2475	-.0297	-.0306	.29	.08	-.4455	-.3948	.0279	-.0590
0	-.06	-.10	.0905	.0909	-.0572	-.0560	-.16	-.28	-.0272	-.0357	-.0755	-.0758	-.35	-.67	-.1359	-.1614	-.1032	-.1023
2	-.23	-.41	.2588	.2142	-.0860	-.0799	-.48	-.82	.2163	.1759	-.1052	-.0974	-.76	-1.30	.1708	.0897	-.1384	-.1197
4	-.37	-.69	.4102	.3917	-.1068	-.1021	-.71	-1.30	.4320	.3842	-.1327	-.1253	-.99	-1.82	.3663	.2863	-.1217	-.1130
8	-.67	-1.26	.7787	.7415	-.1449	-.1339	-1.24	-2.33	.9046	.8472	-.1976	-.1973	-1.62	-3.04	.7837	.6683	-.1589	-.1427
12	-.88	-1.66	1.1347	1.0689	-.2329	-.2051	-1.58	-2.96	1.0987	1.0295	-.2167	-.2012	-2.04	-3.86	1.0017	.9260	-.1958	-.1856
16	-.78		1.0771		-.2197		-1.32		.8879		-.1809		-1.65		.6916		-.1293	
20	-.83		1.1576		-.2322		-1.42		.9257		-.1953		-1.78		.7678		-.1463	
M = 1.200																		
-4	0.26	0.54	-0.2589	-0.2702	0.0033	-0.0165	0.47	1.01	-0.4246	-0.4610	0.0127	0.0151	0.60	1.37	-0.5739	-0.6485	0.0556	0.0824
-2	.15	.21	-.1420	-.1269	-.0270	-.0587	.23	.51	-.2965	-.2593	-.0187	-.0340	.27	.16	-.4518	-.3849	-.0032	-.0491
0	.00	-.08	.0221	.0550	-.0534	-.0601	-.02	-.21	-.1059	-.0534	-.0518	-.0589	-.13	-.50	-.2602	-.1767	-.0576	-.0774
2	-.18	-.38	.2023	.2111	-.0826	-.0827	-.36	-.74	.1403	.1463	-.0886	-.0867	-.58	-1.18	.0298	.0426	-.1033	-.1053
4	-.34	-.66	.3751	.3751	-.1038	-.1036	-.69	-1.25	.3569	.3405	-.1193	-.1164	-.91	-1.77	.2840	.2519	-.1150	-.1122
8	-.62	-1.21	.7162	.7042	-.1508	-.1435	-1.14	-2.23	.7561	.7443	-.1812	-.1846	-1.48	-2.92	.6257	.5889	-.1301	-.1322
12	-.83	-1.63	.9915	.9618	-.2170	-.2019	-1.50	-2.96	.9821	.9774	-.2198	-.2231	-1.92	-3.82	.8254	.8246	-.1599	-.1669
16	-.93		1.1702		-.2371		-1.63		1.0863		-.2250		-2.09		.9201		-.1843	
20	-.96		1.2192		-.2603		-1.64		1.0894		-.2303		-2.09		.7909		-.1592	

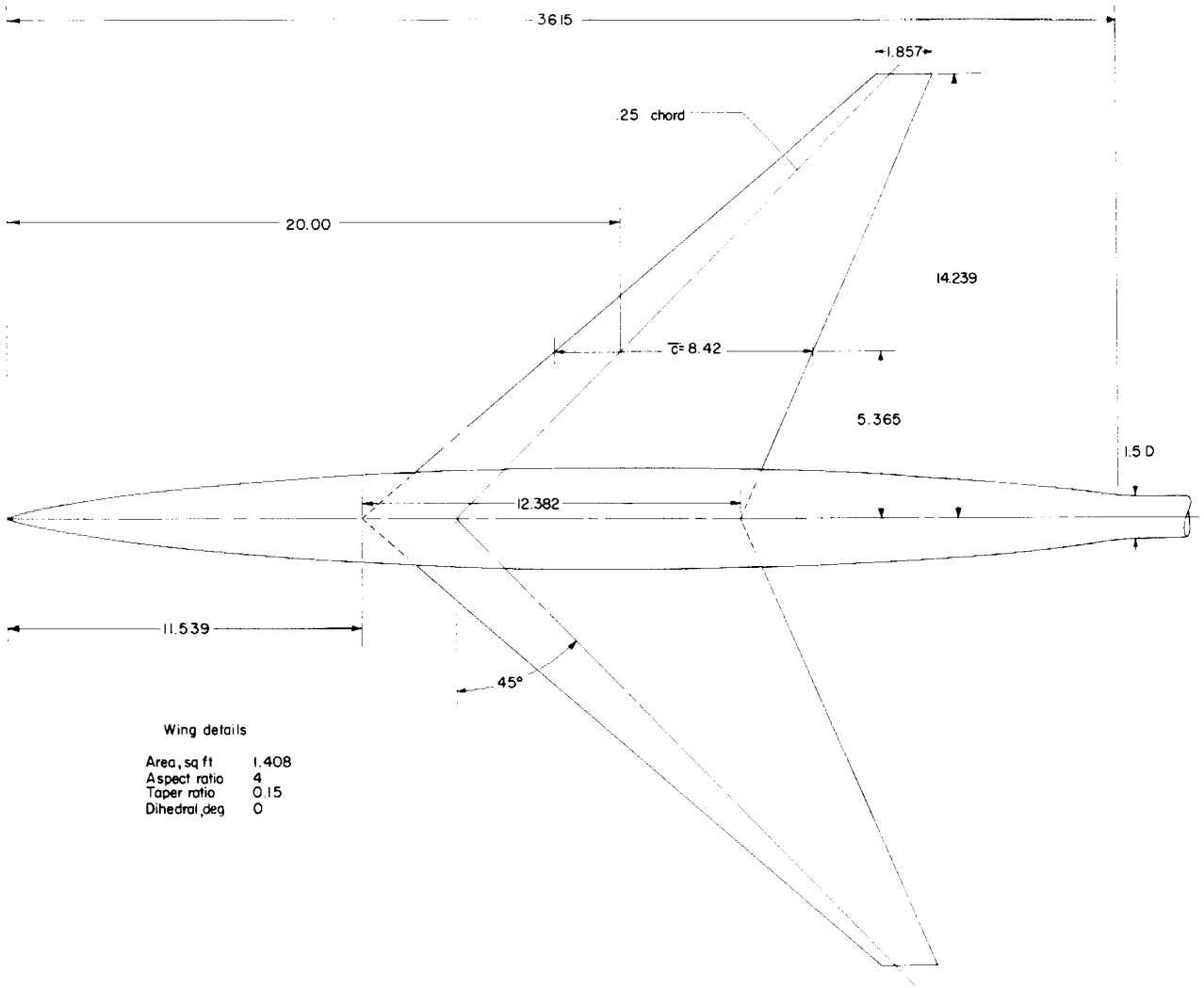


Figure 1.- Details of wing-body combination. All dimensions are in inches unless otherwise noted.

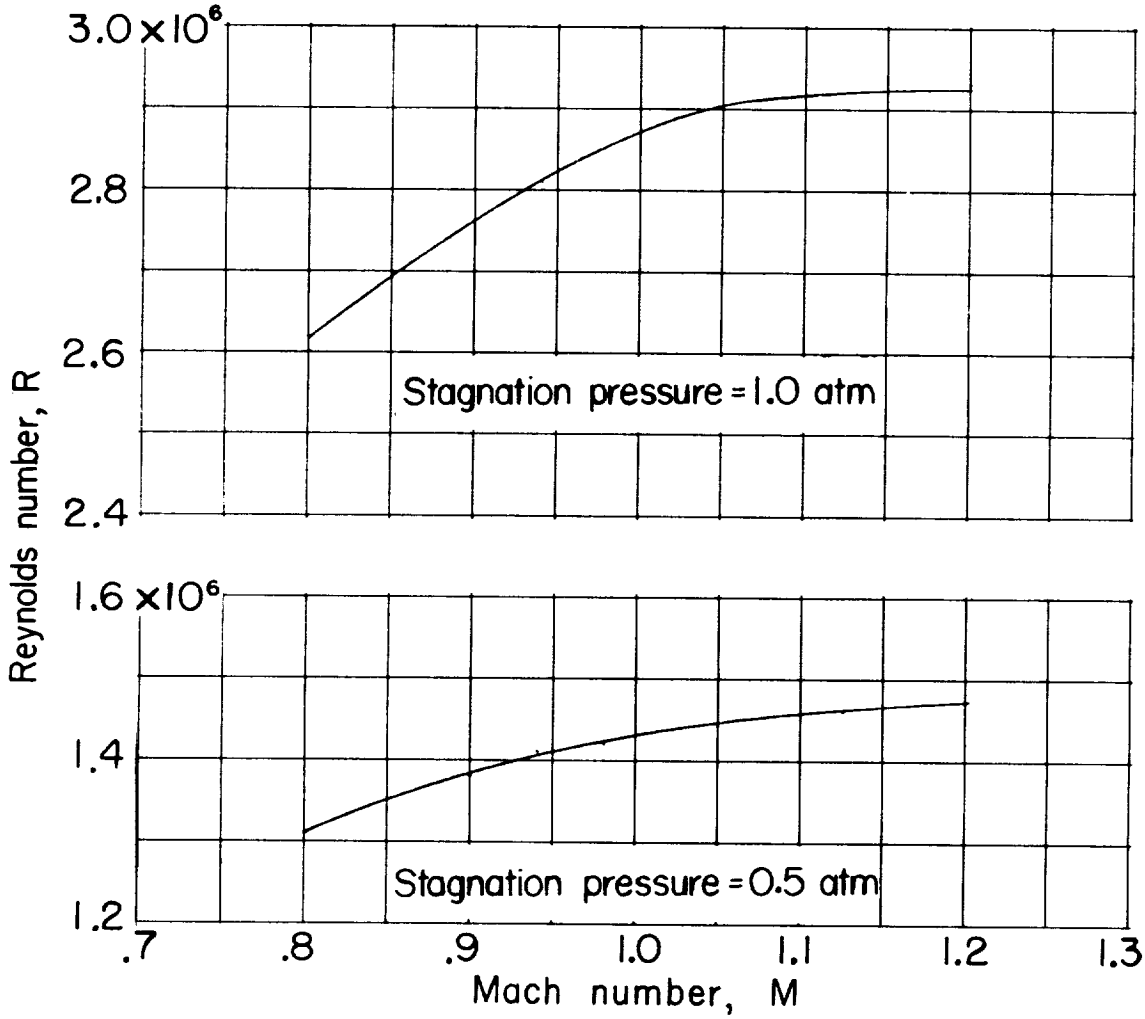


Figure 2.- Variation with Mach number of average Reynolds number based on wing mean aerodynamic chord.

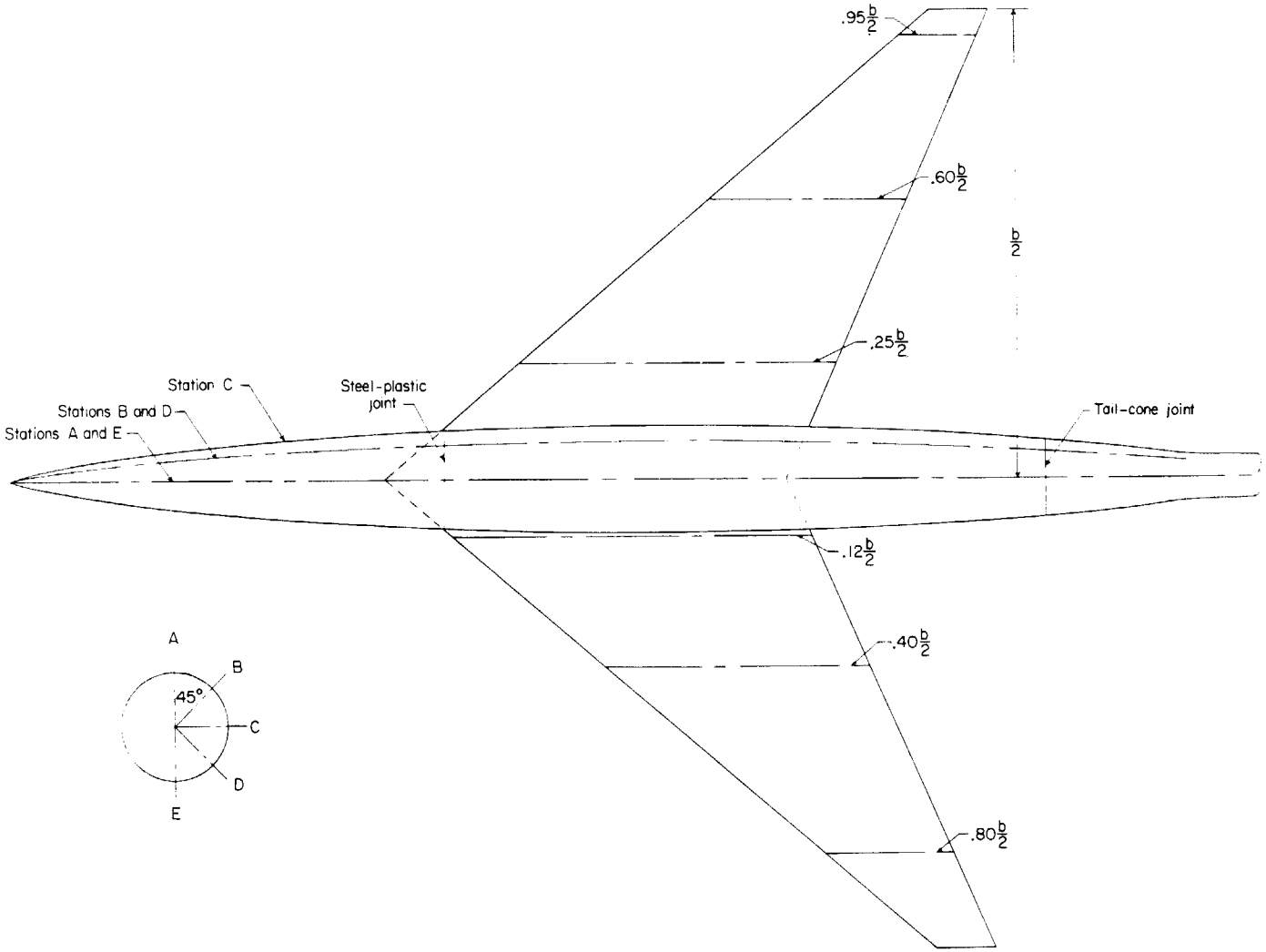
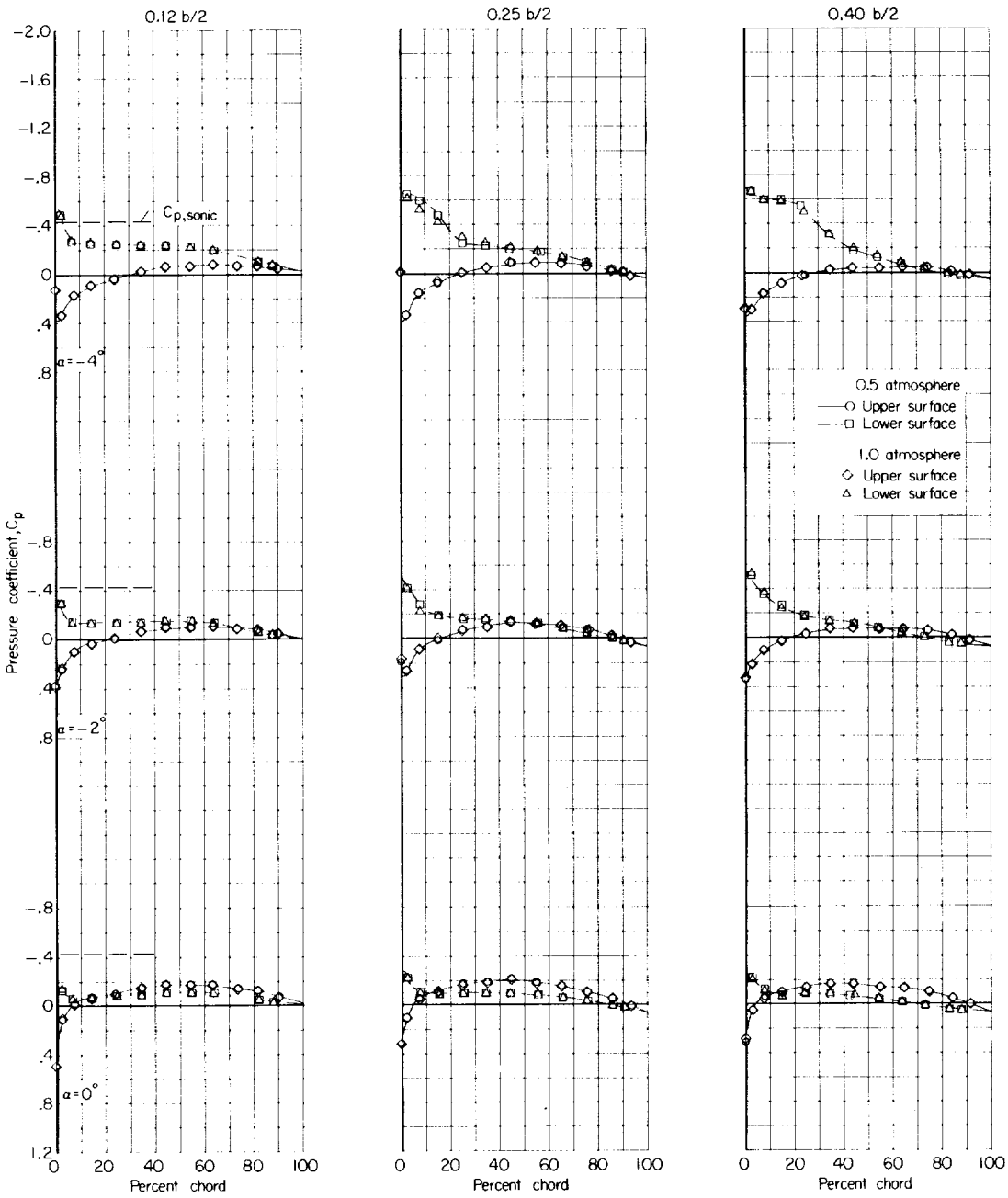
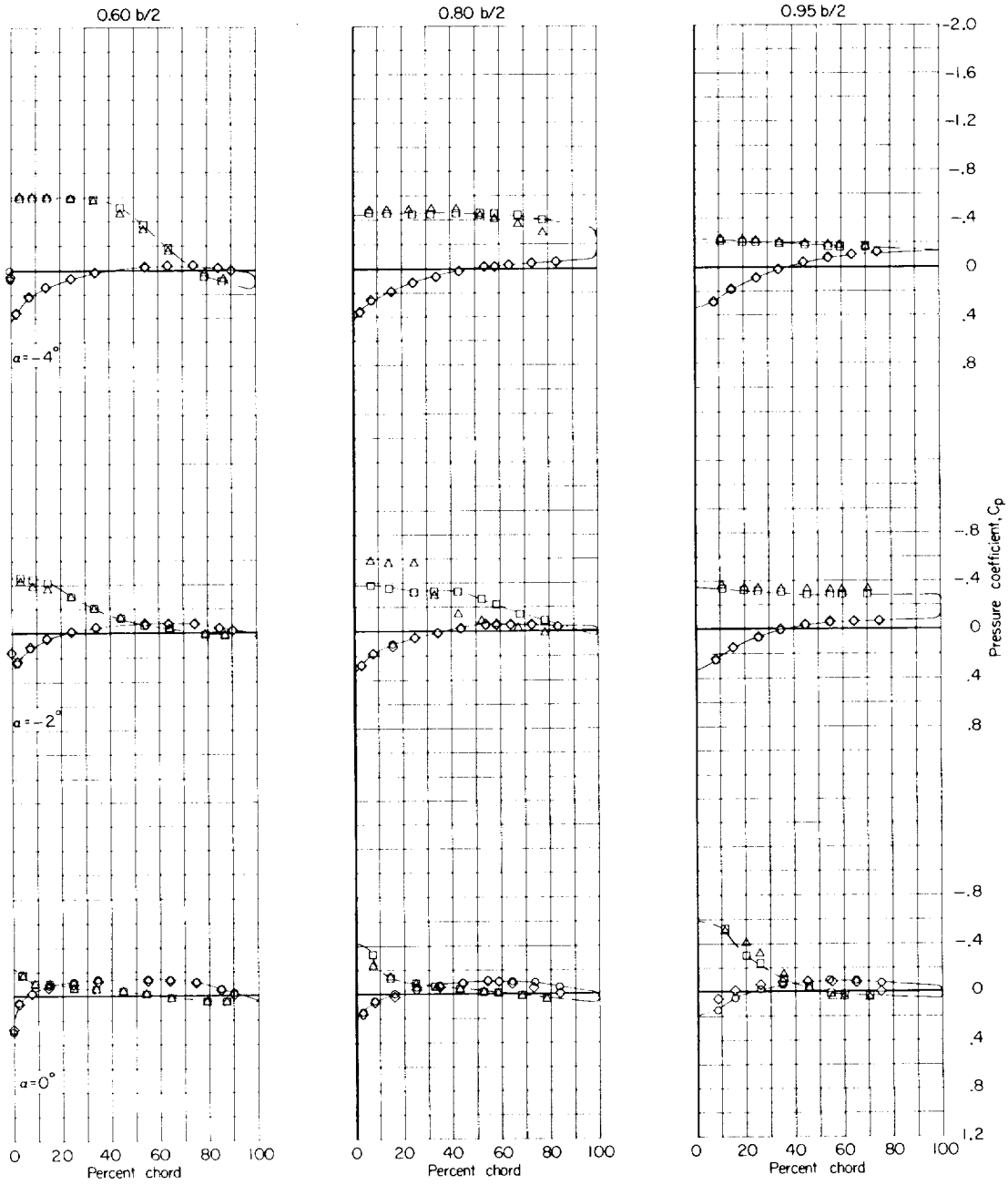


Figure 3.- Sketch of the location of pressure orifices on the wing and body.



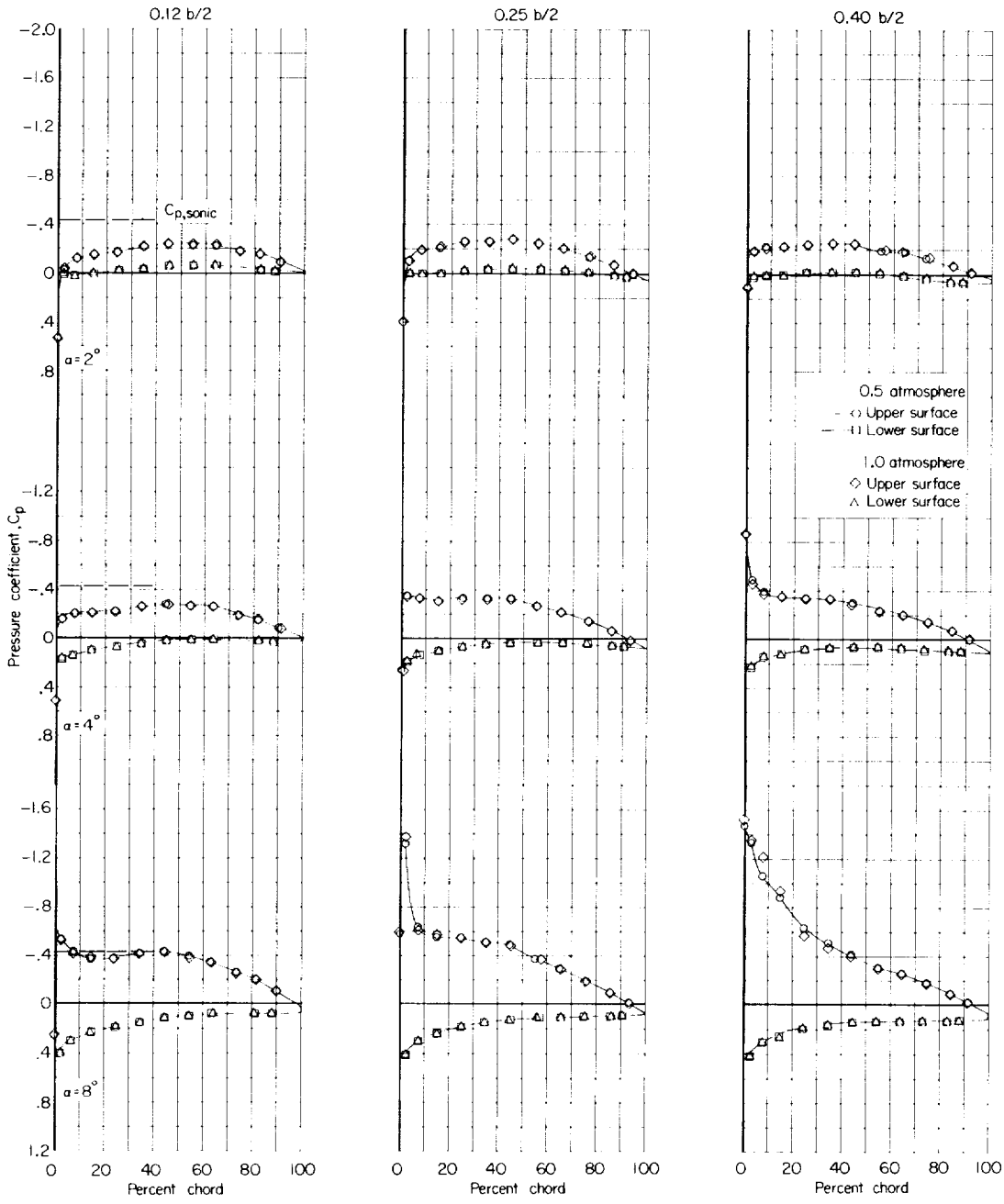
(a) M = 0.800; α = -4°, -2°, and 0°.

Figure 4.- Pressure measurements on the wing in the presence of the body.



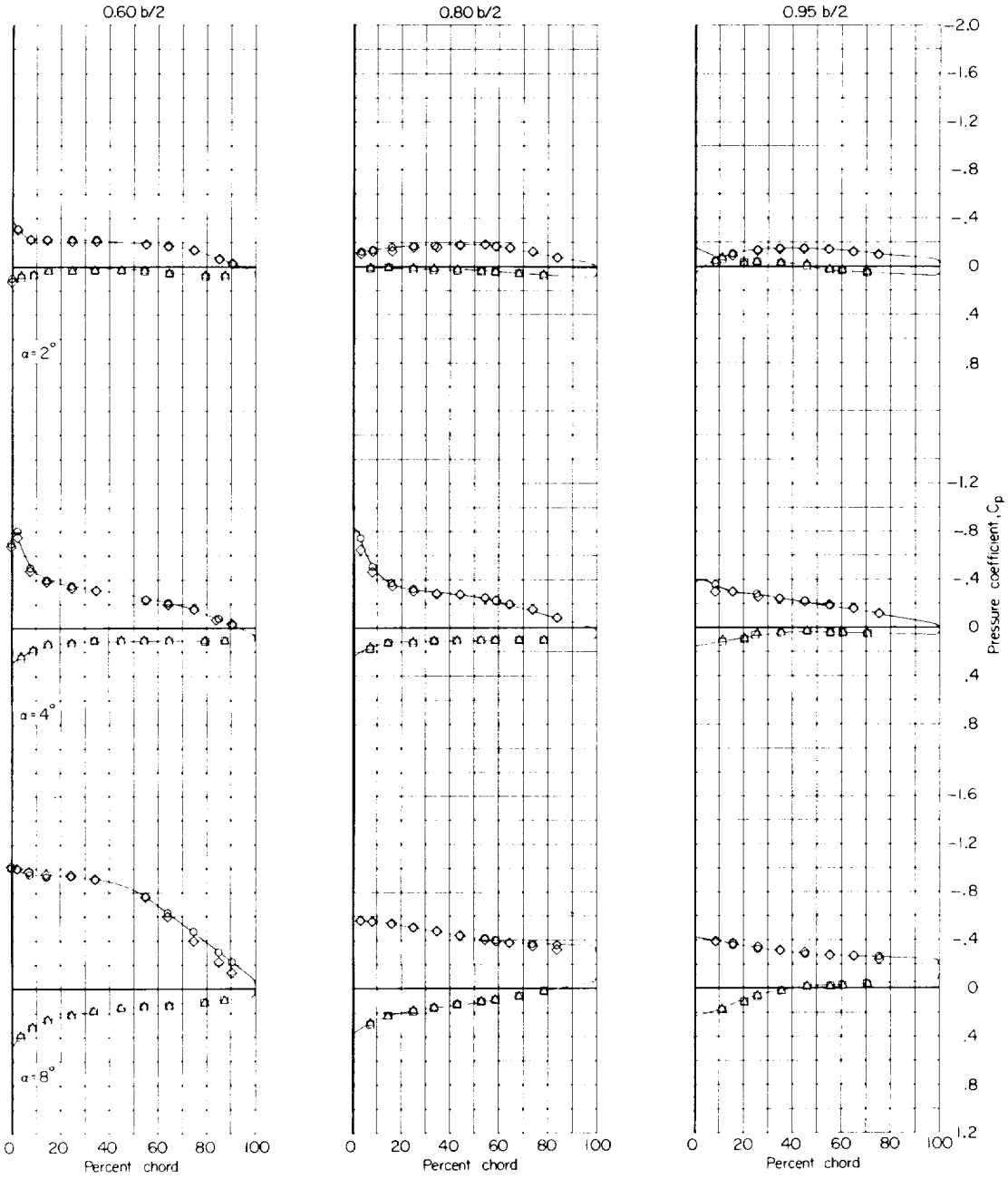
(a) Concluded.

Figure 4.- Continued.



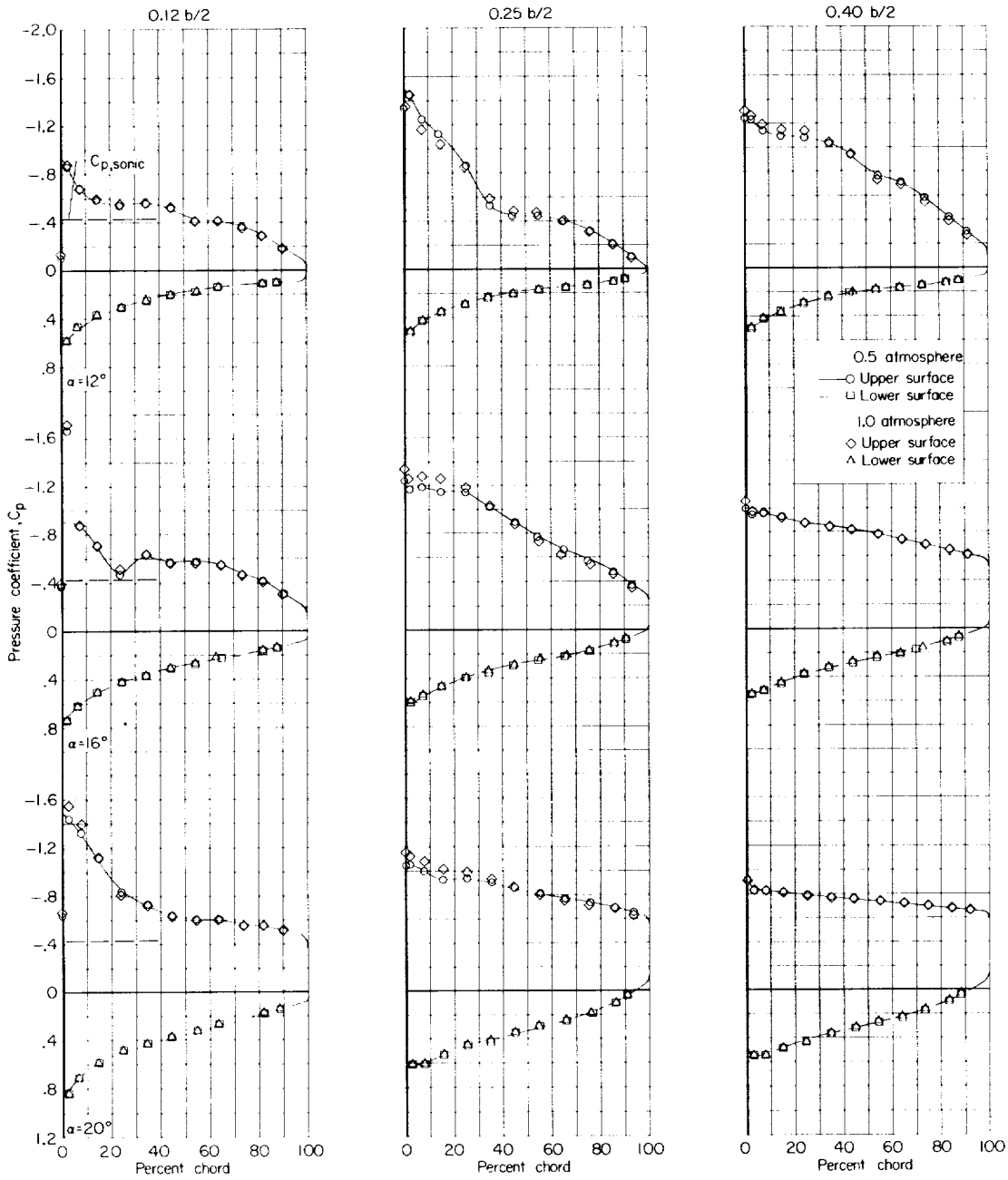
(b) M = 0.800; α = 2°, 4°, and 8°.

Figure 4.- Continued.



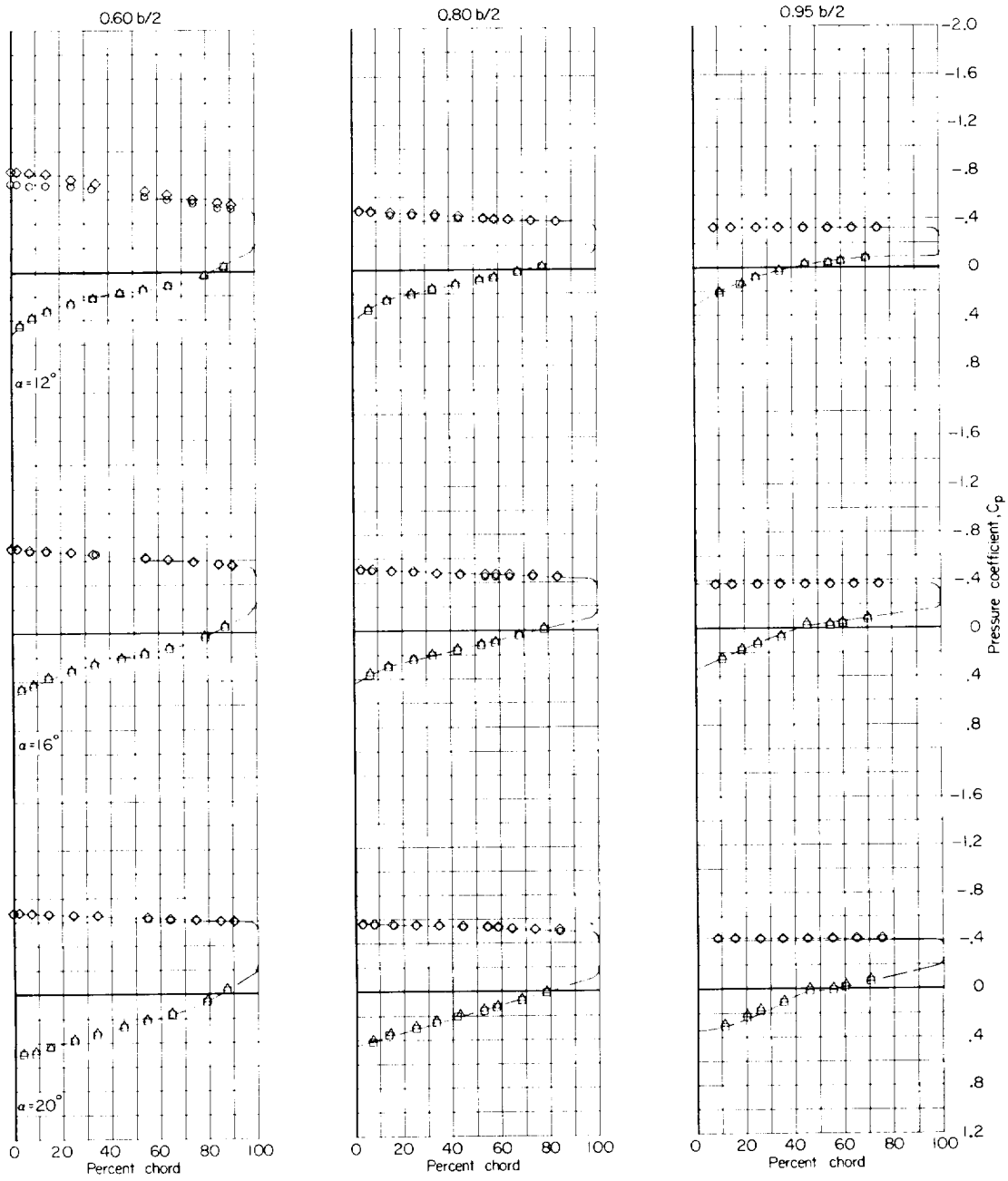
(b) Concluded.

Figure 4.- Continued.



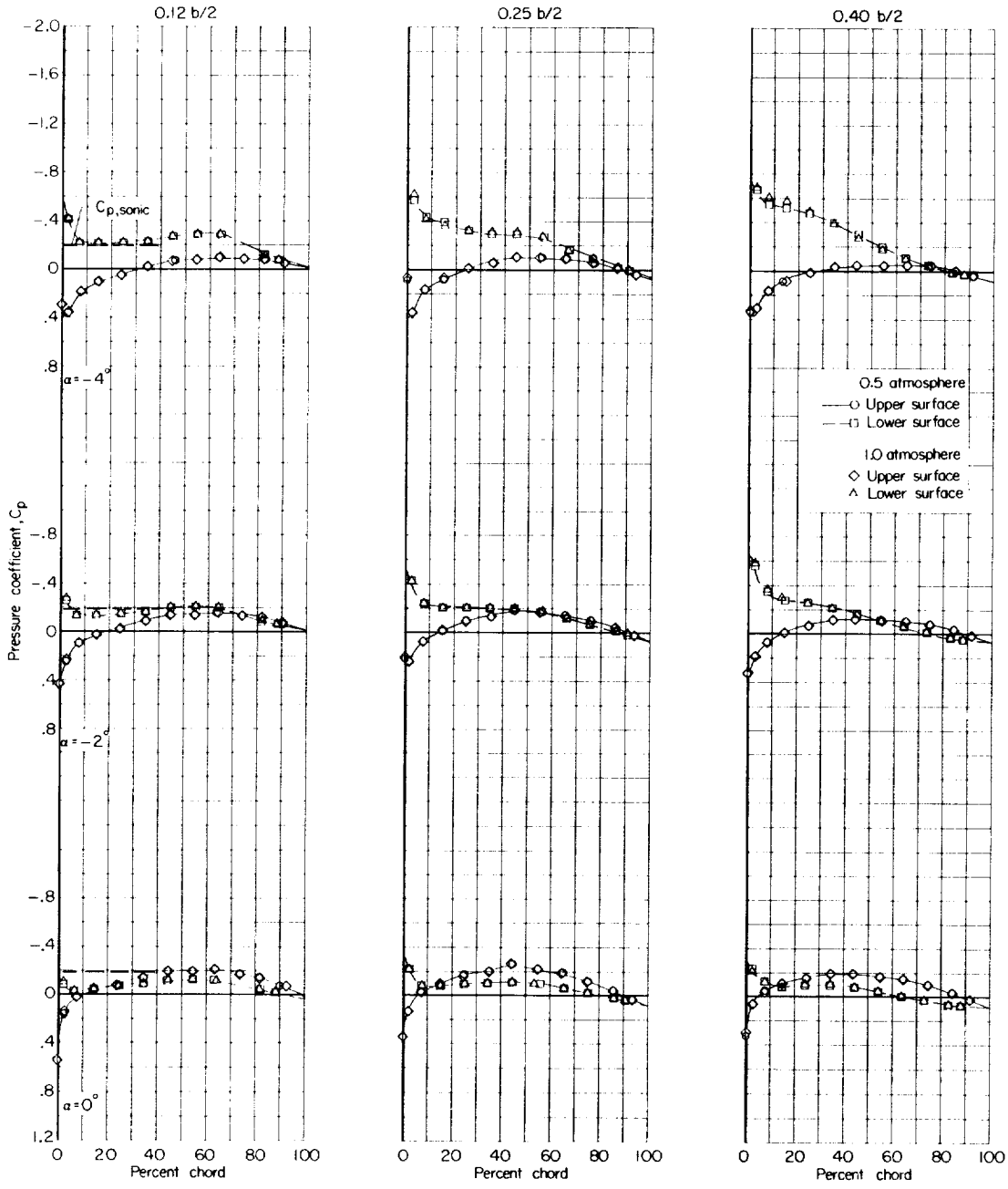
(c) M = 0.800; α = 12°, 16°, and 20°.

Figure 4.- Continued.



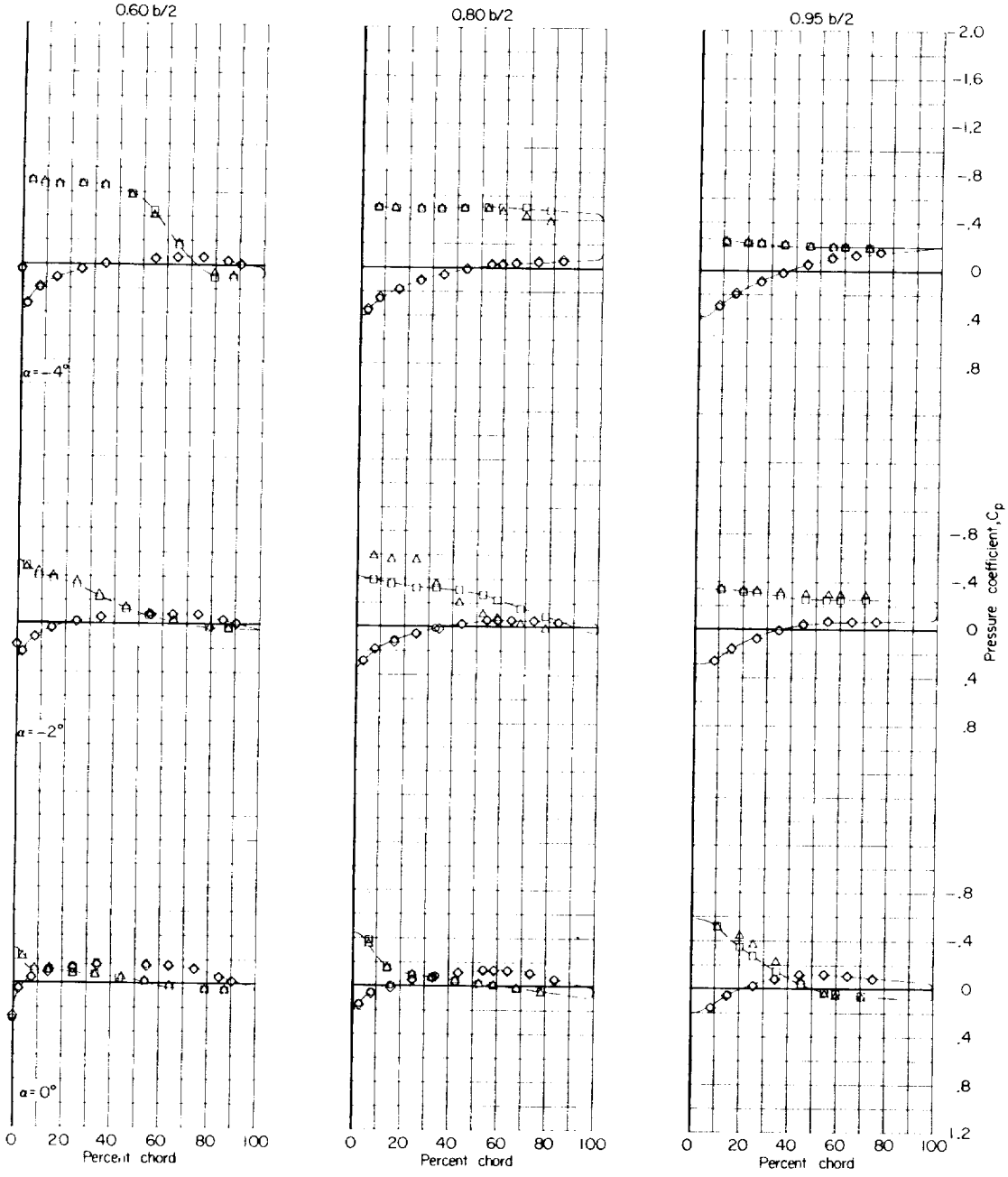
(c) Concluded.

Figure 4.- Continued.



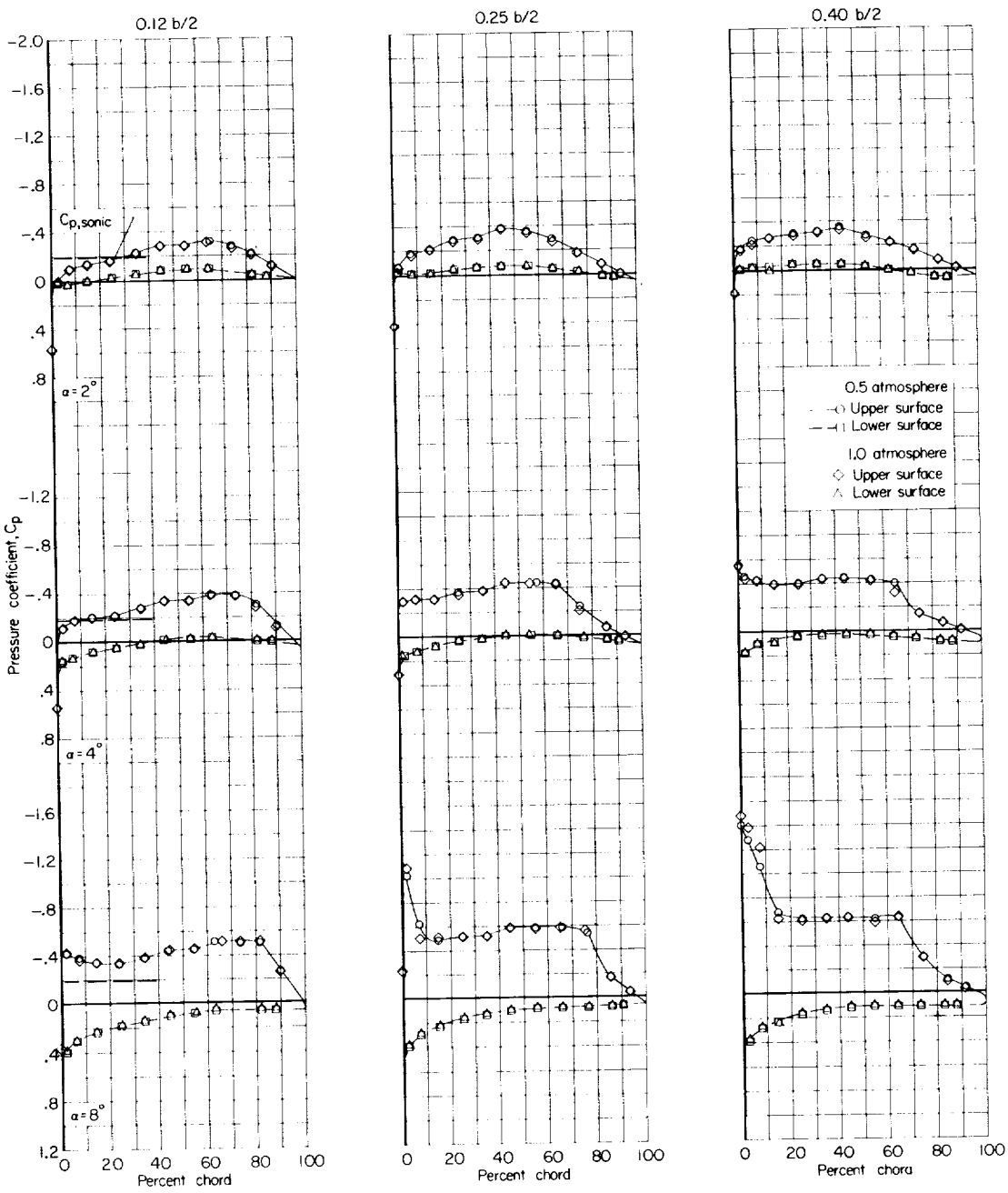
(d) M = 0.900; α = -4°, -2°, and 0°.

Figure 4.- Continued.



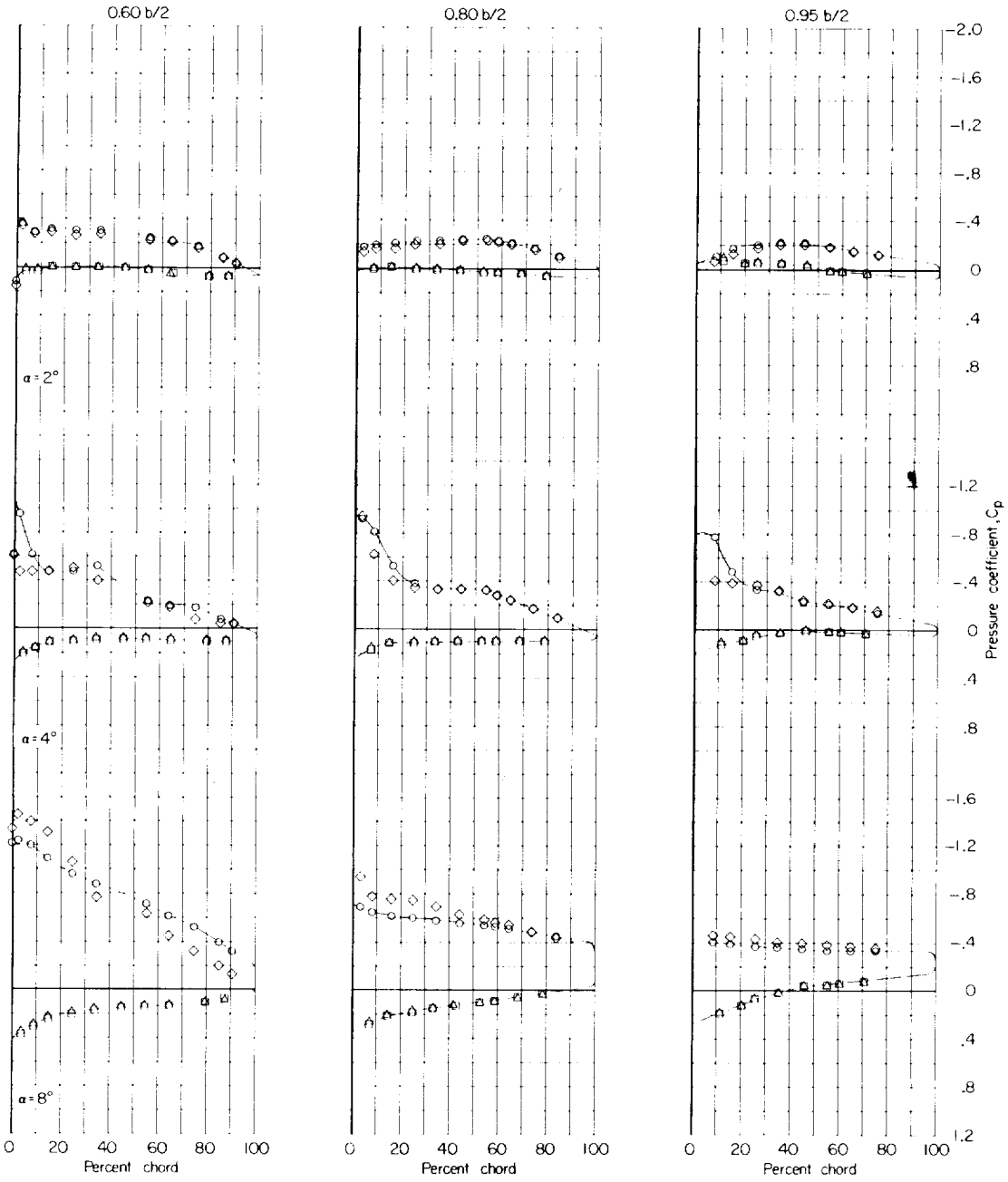
(d) Concluded.

Figure 4.- Continued.



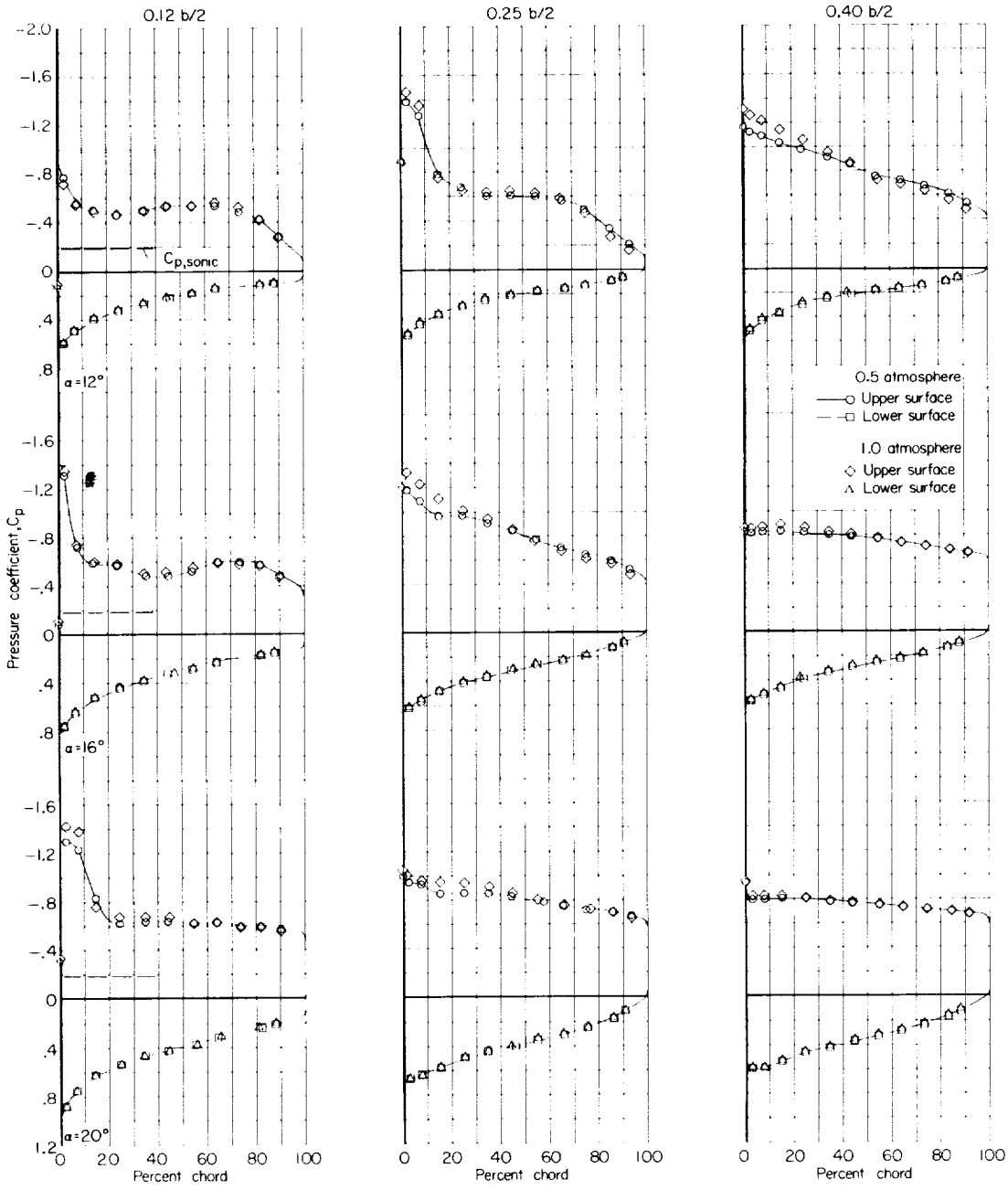
(e) M = 0.900; α = 2°, 4°, and 8°.

Figure 4.- Continued.



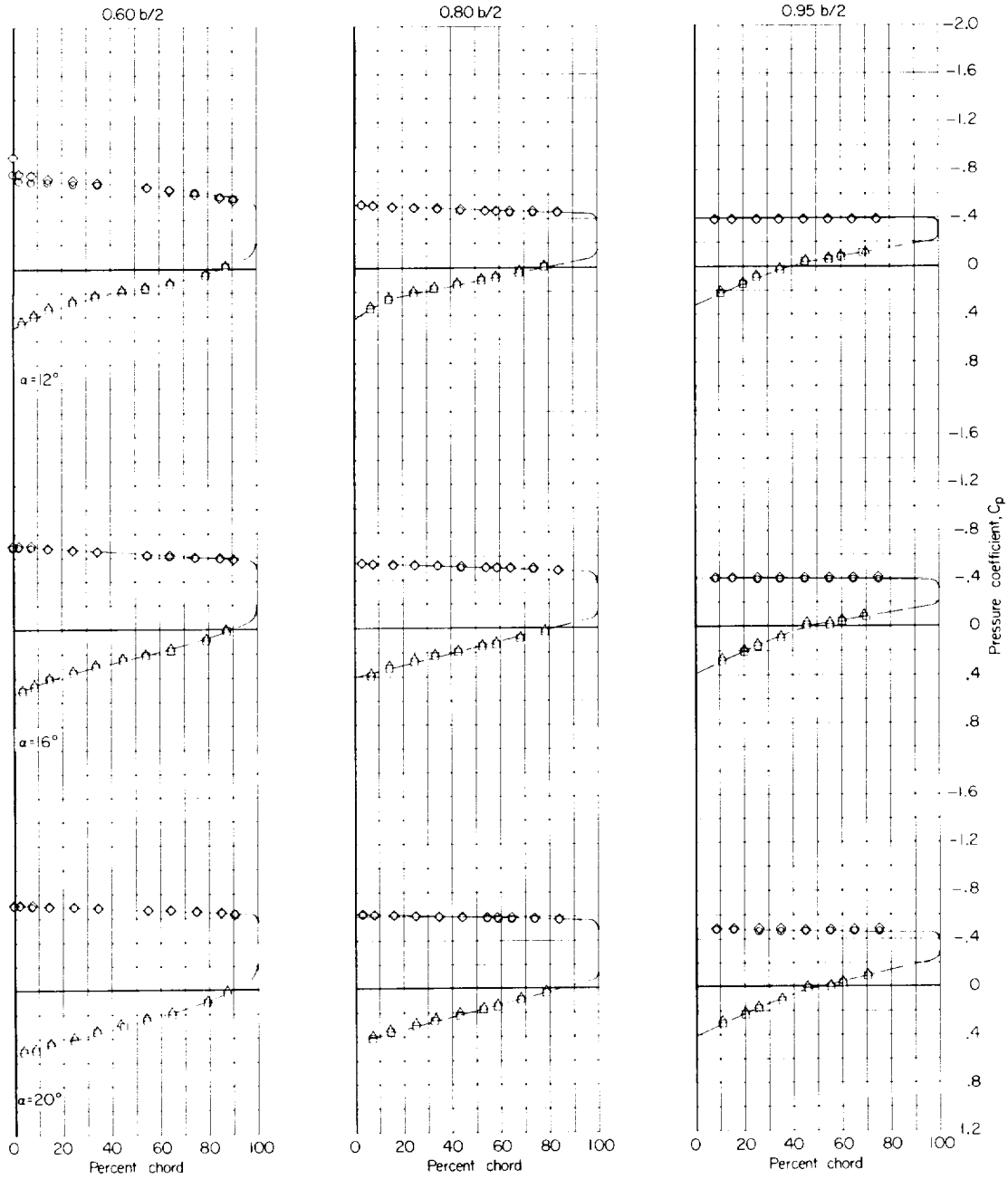
(e) Concluded.

Figure 4.- Continued.



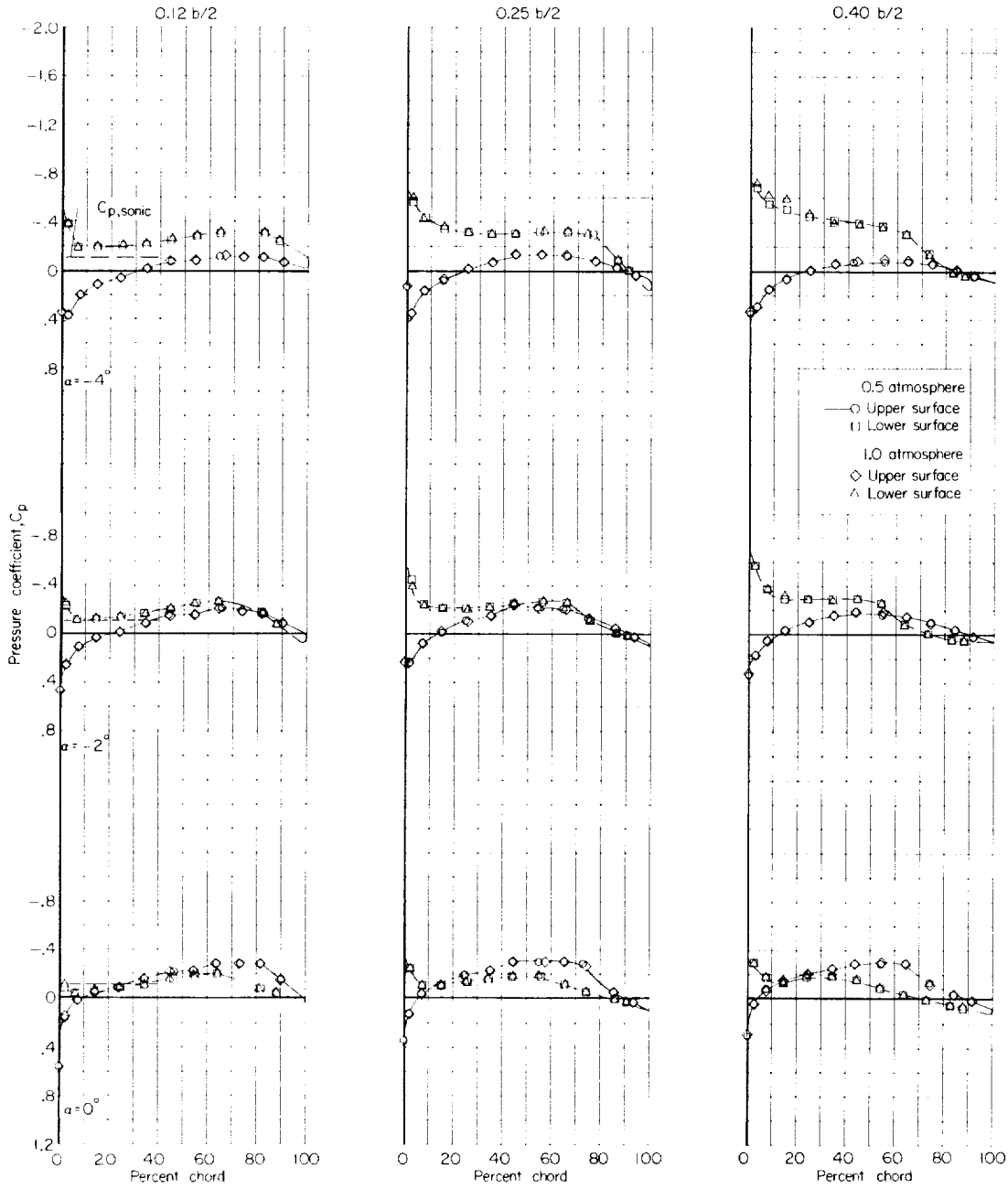
(f) M = 0.900; α = 12°, 16°, and 20°.

Figure 4.- Continued.



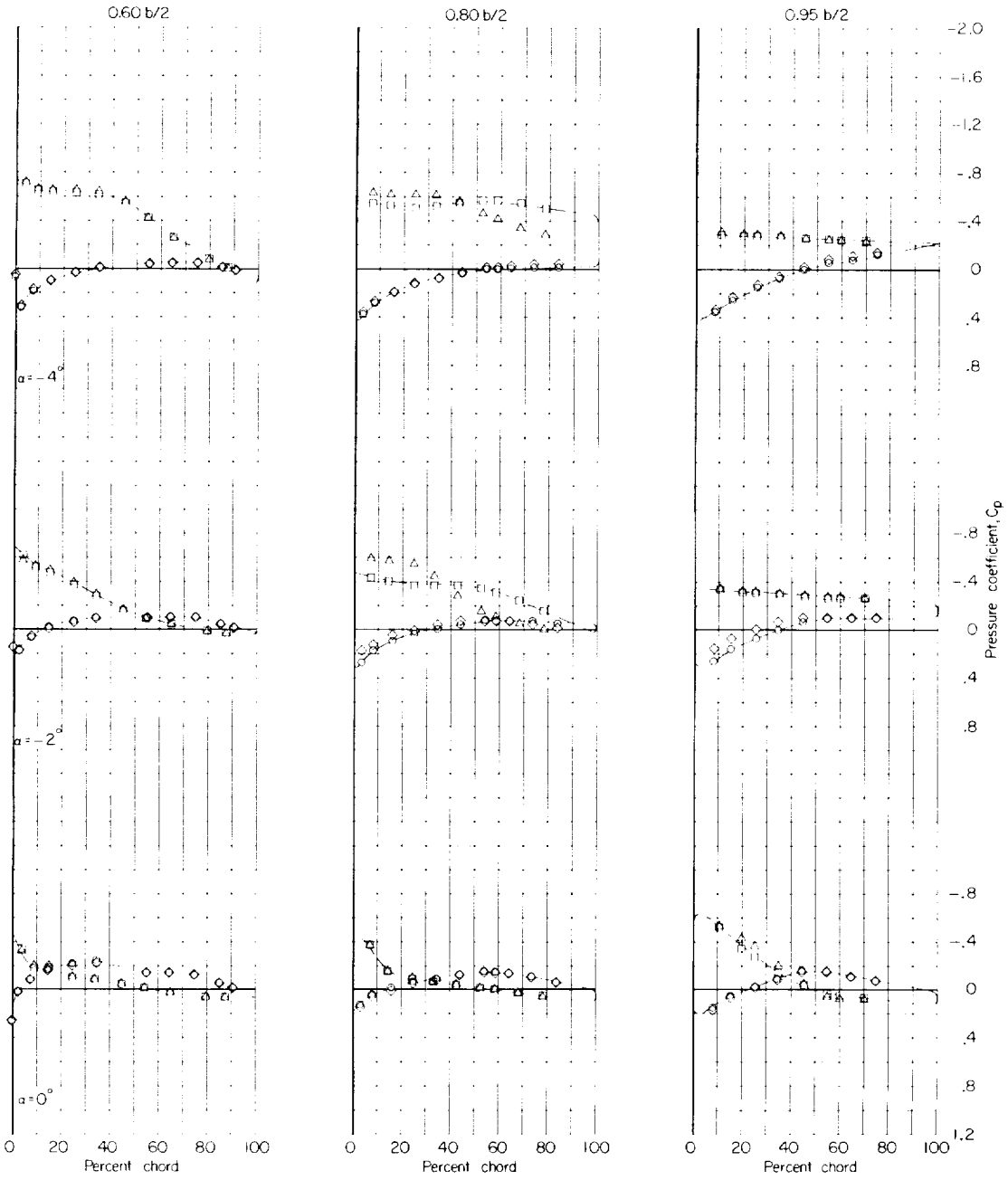
(f) Concluded.

Figure 4.- Continued.



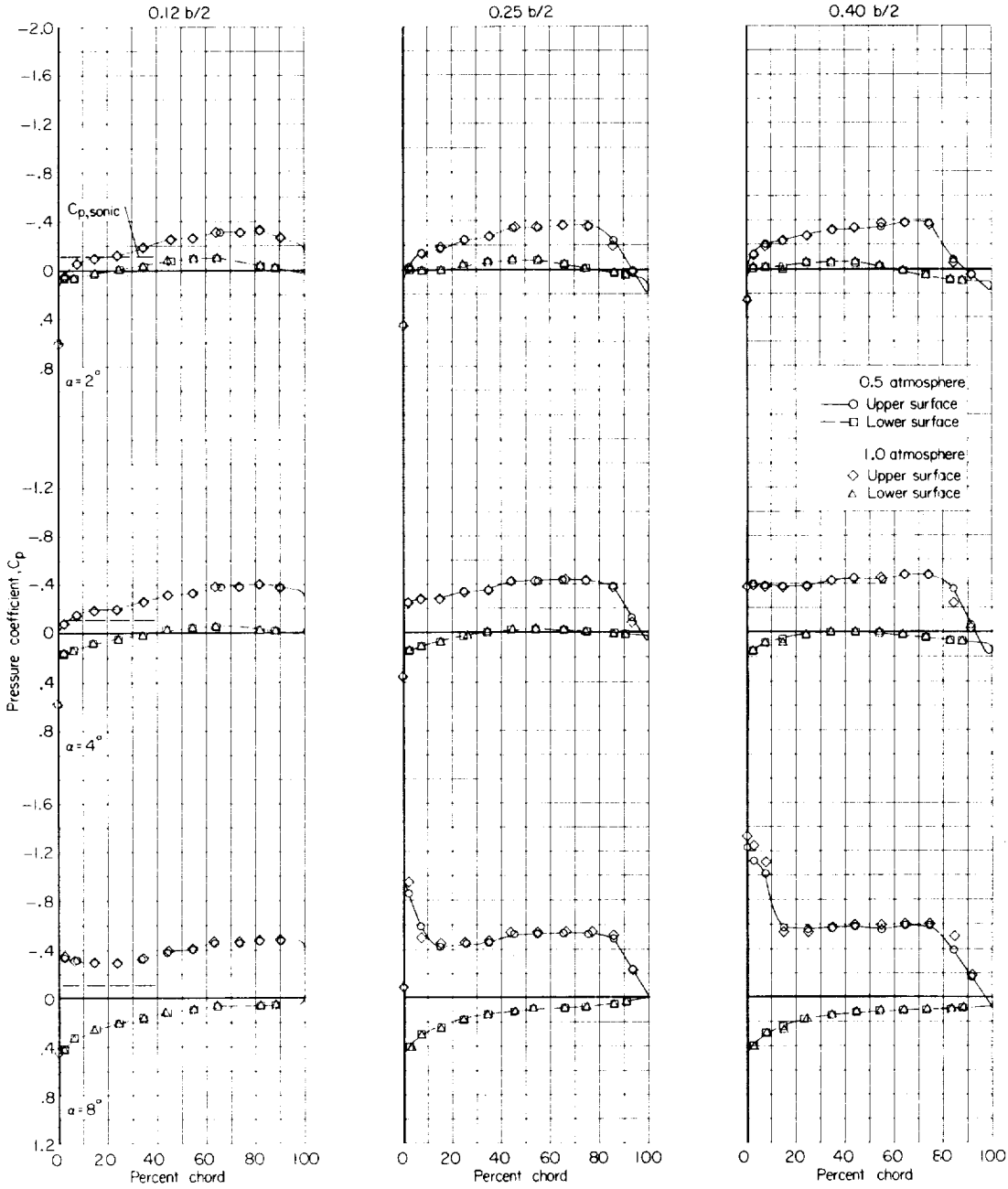
(g) M = 0.940; α = -4°, -2°, and 0°.

Figure 4.- Continued.



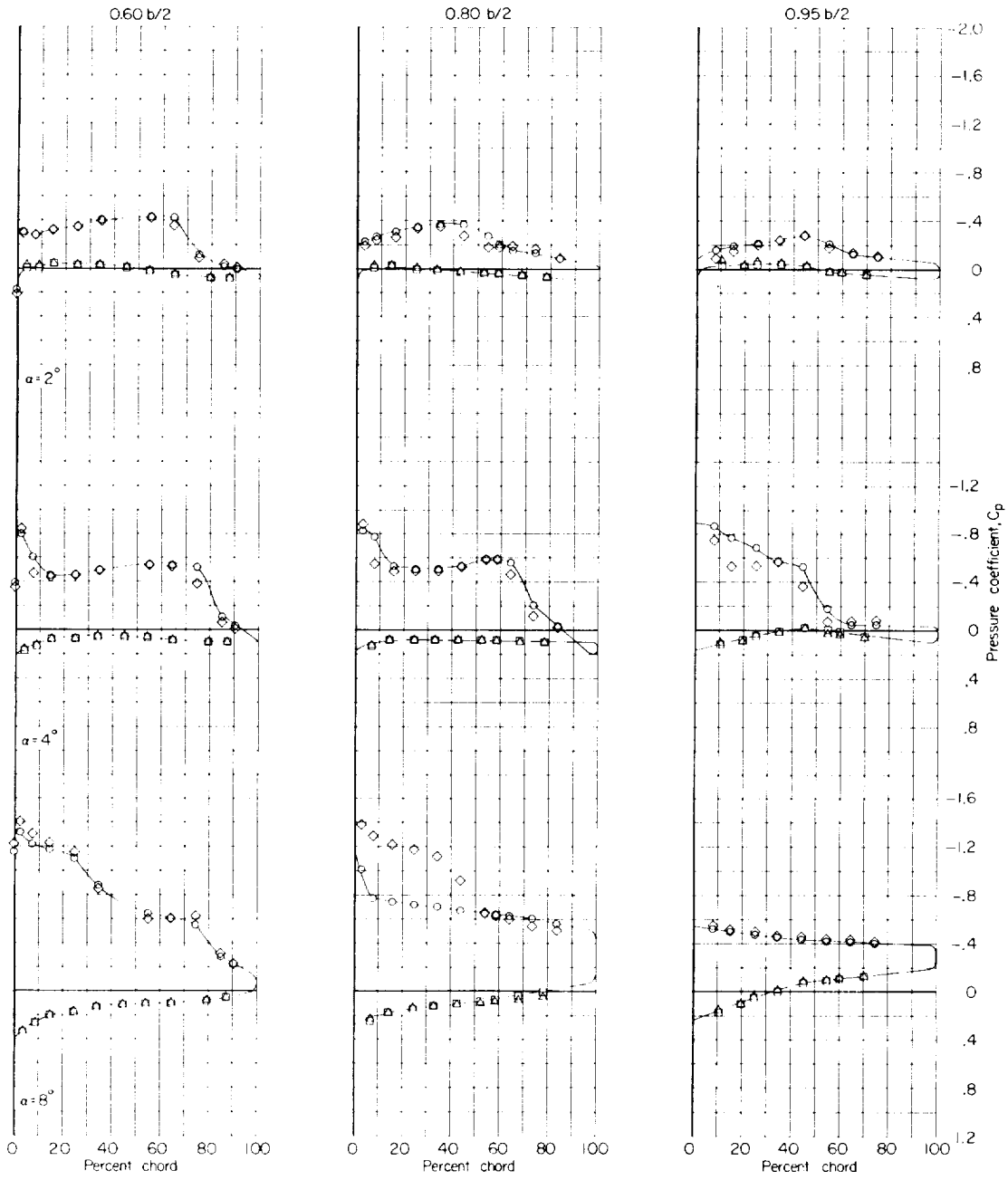
(g) Concluded.

Figure 4.- Continued.



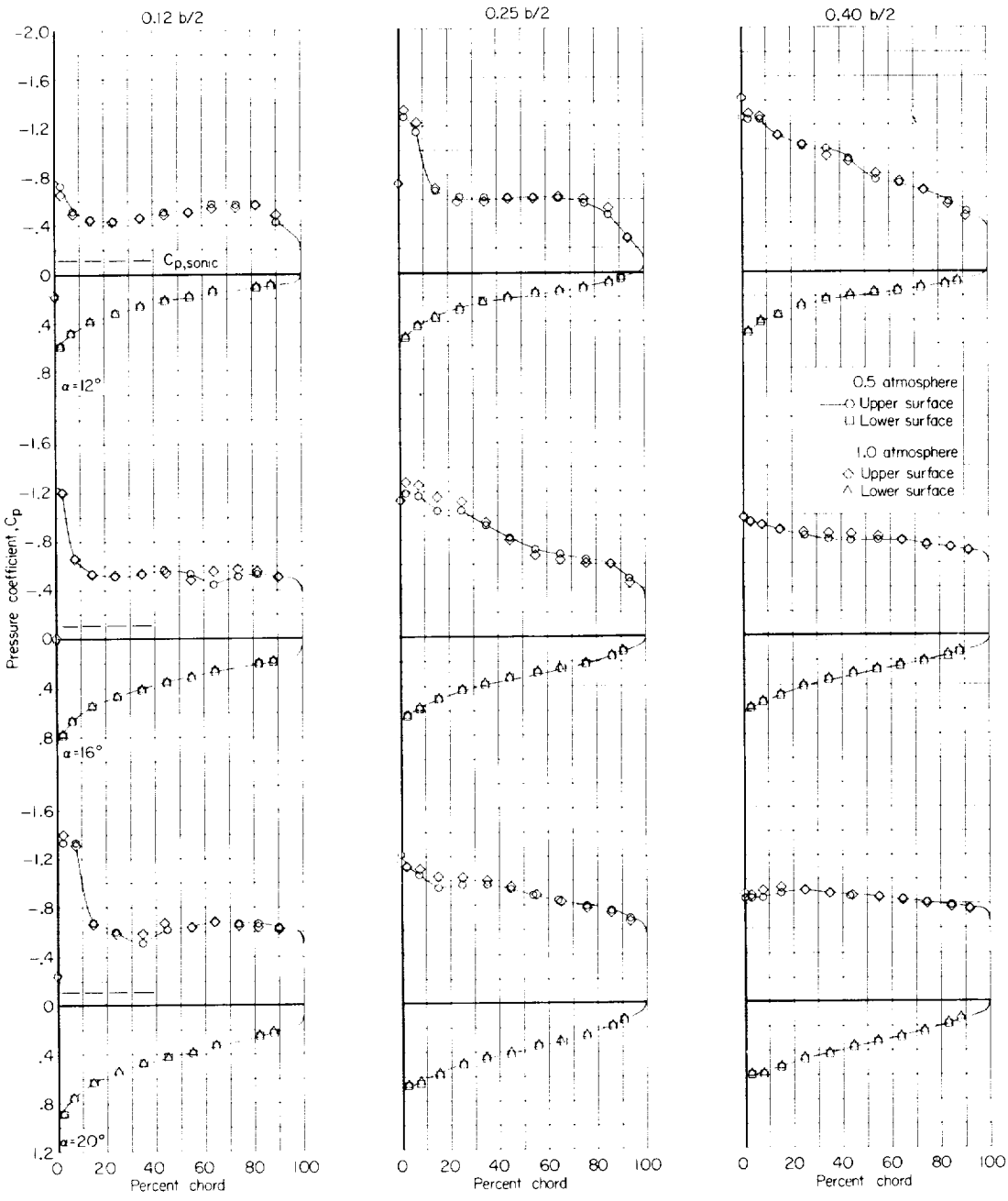
(h) M = 0.940; α = 2°, 4°, and 8°.

Figure 4.- Continued.



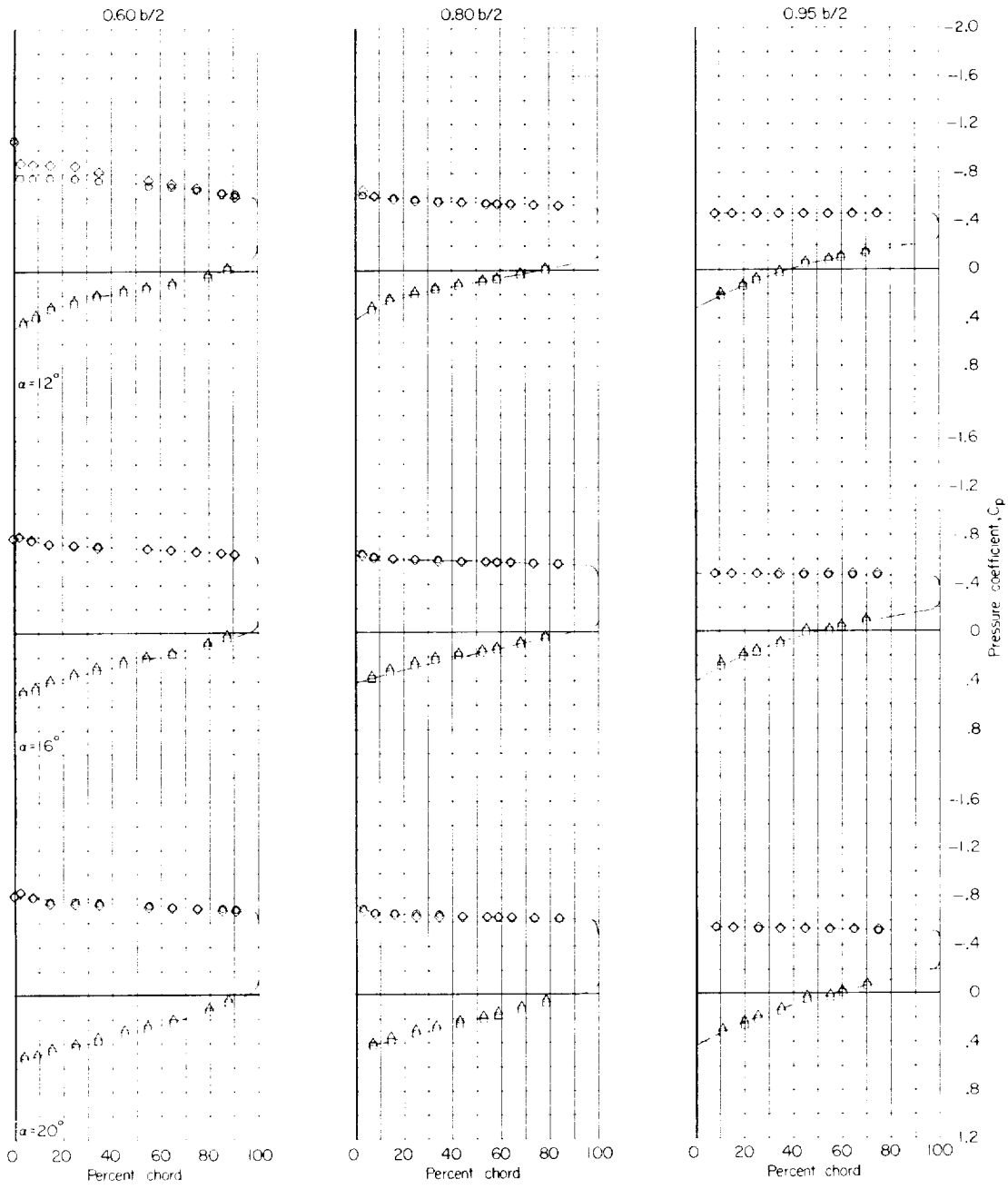
(h) Concluded.

Figure 4.- Continued.



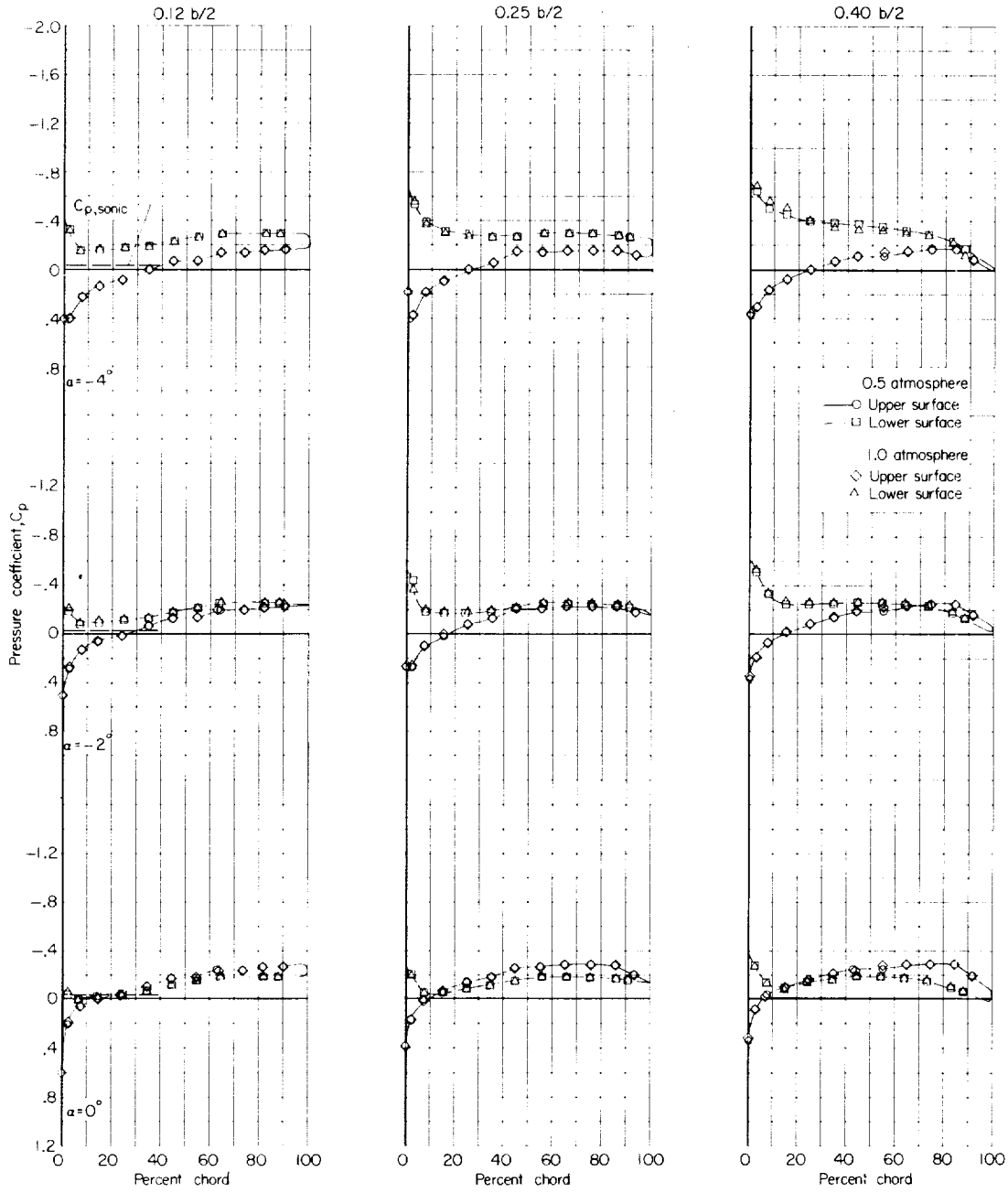
(i) M = 0.940; α = 12°, 16°, and 20°.

Figure 4.- Continued.



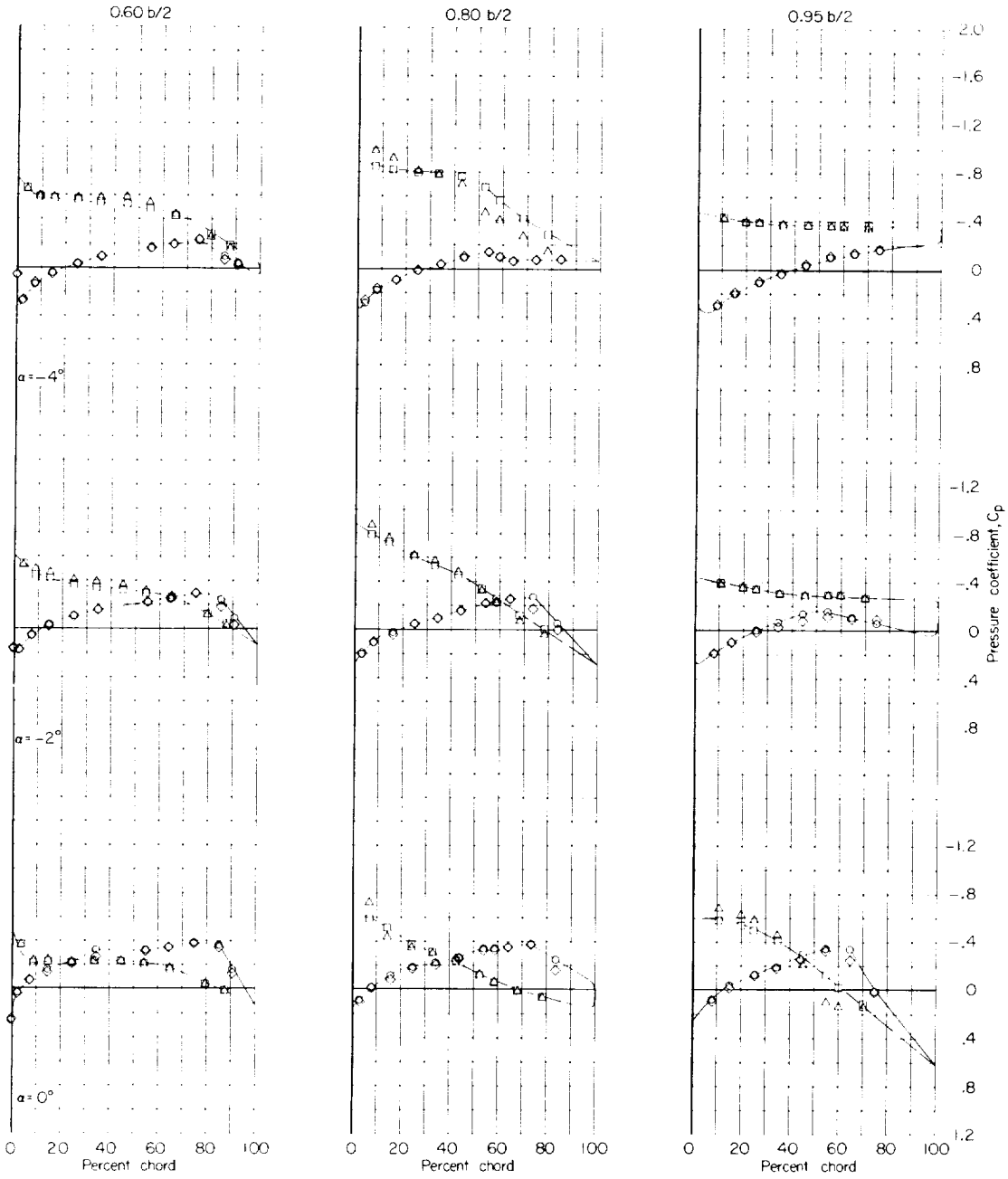
(i) Concluded.

Figure 4.- Continued.



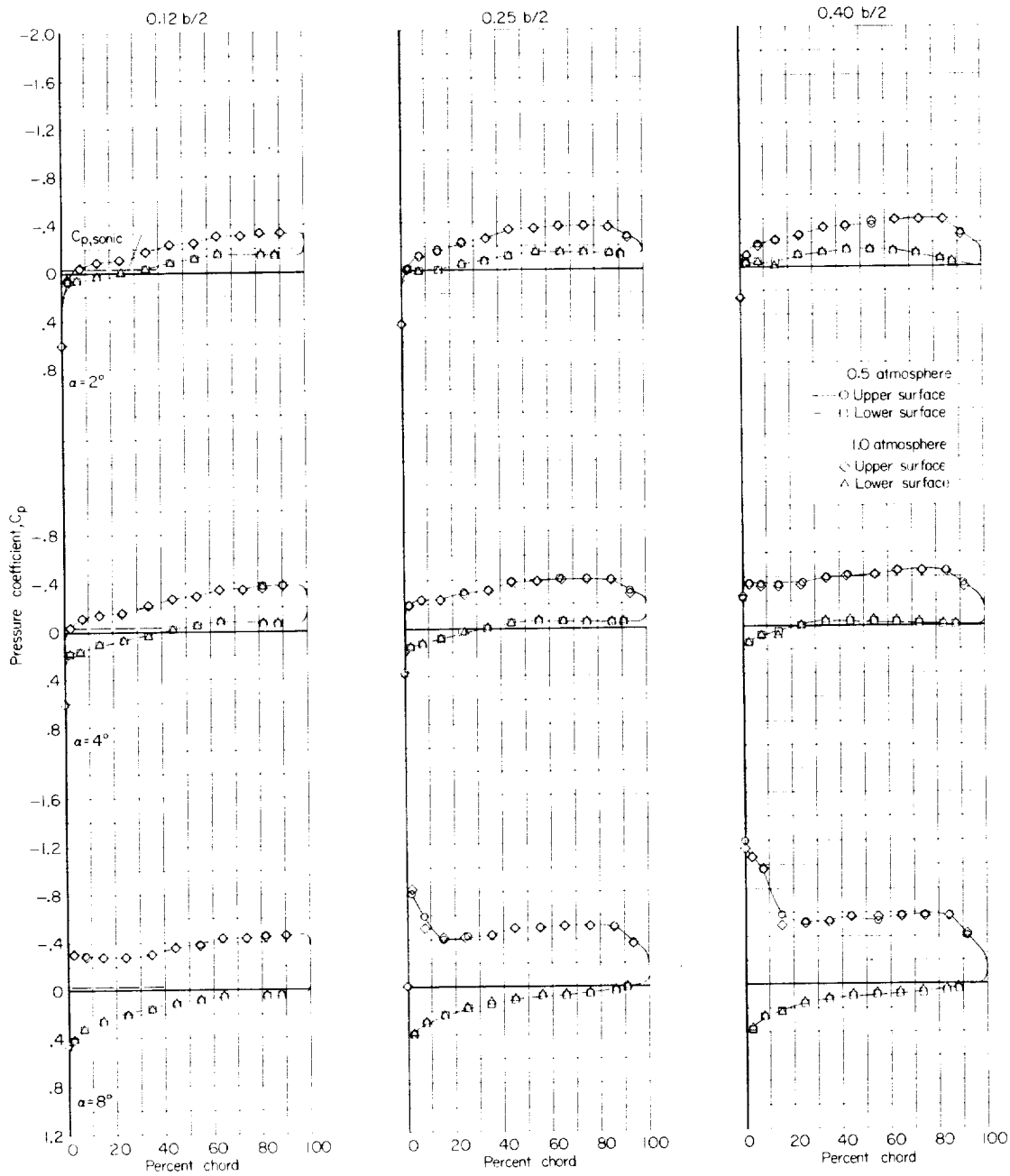
(j) M = 0.980; α = -4°, -2°, and 0°.

Figure 4.- Continued.



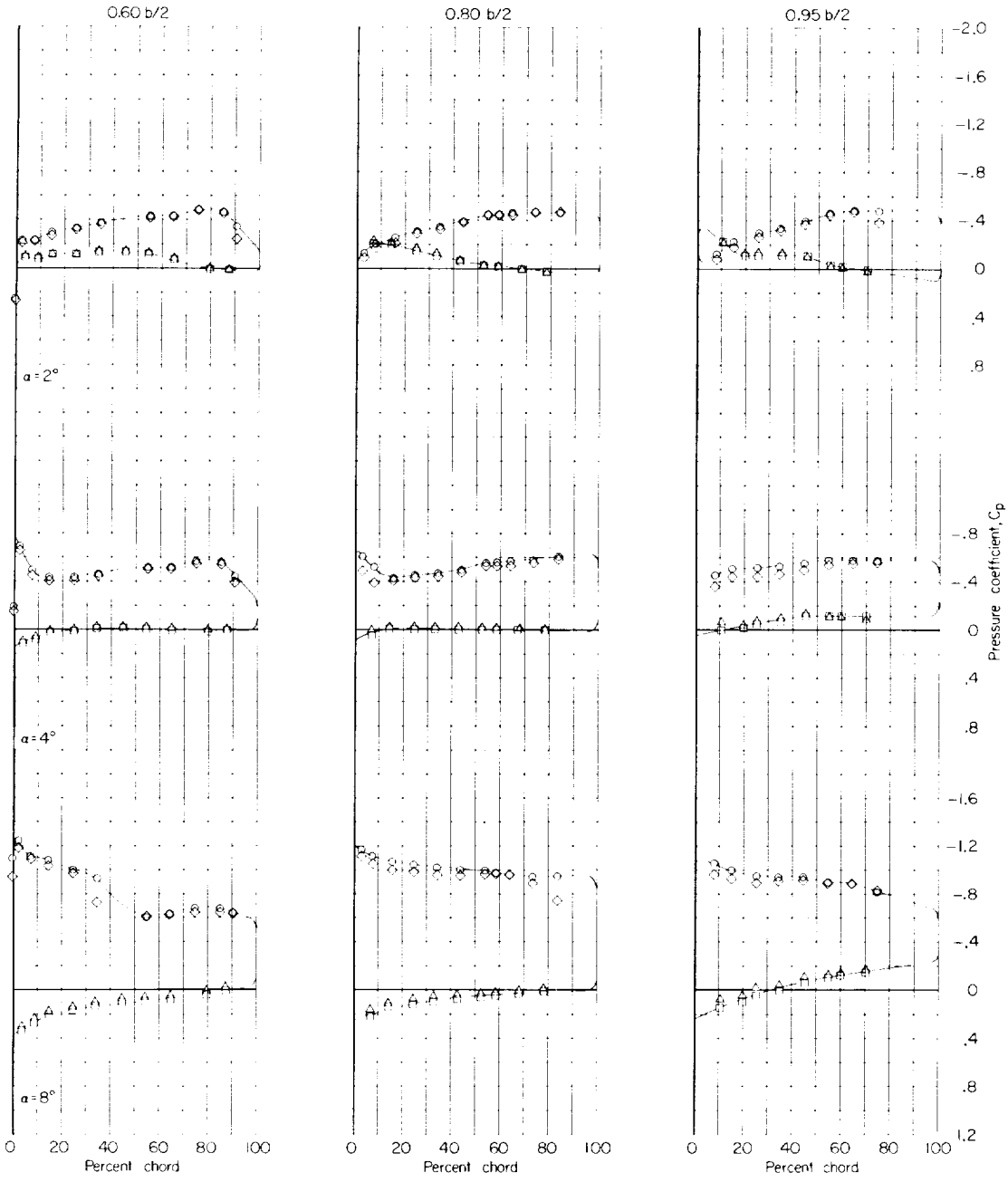
(j) Concluded.

Figure 4.- Continued.



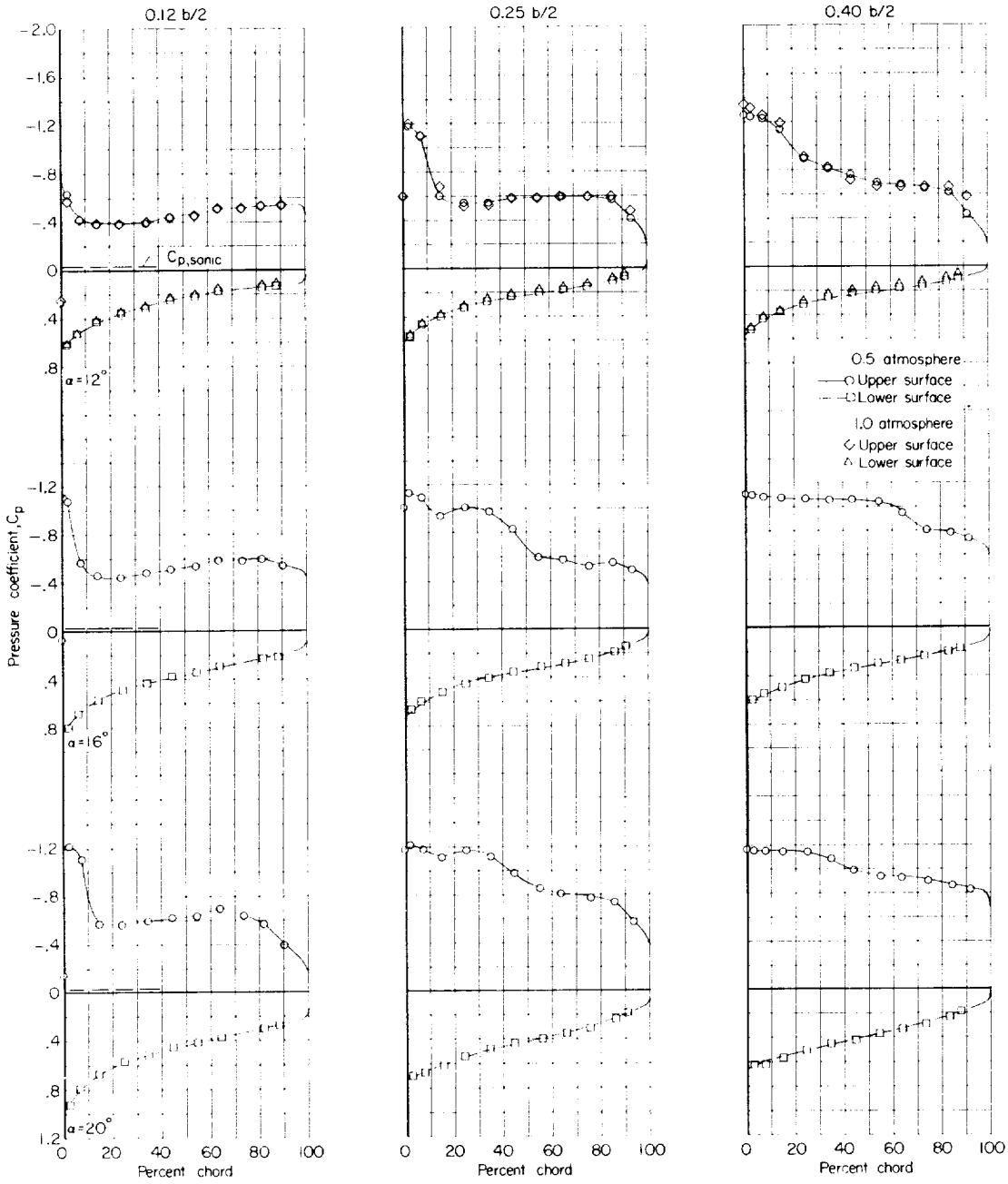
(k) M = 0.980; $\alpha = 2^{\circ}$, 4°, and 8°.

Figure 4.- Continued.



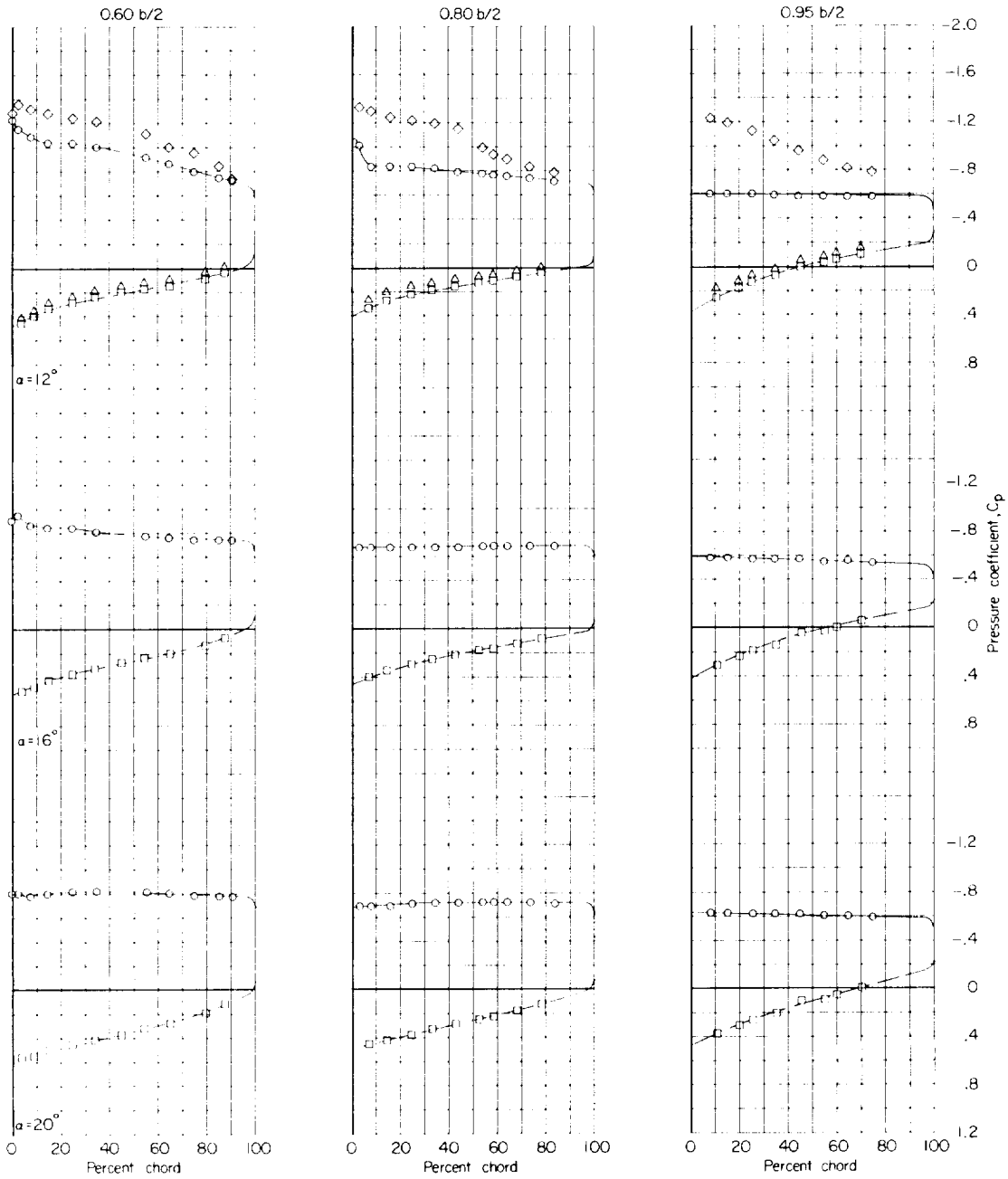
(k) Concluded.

Figure 4.- Continued.



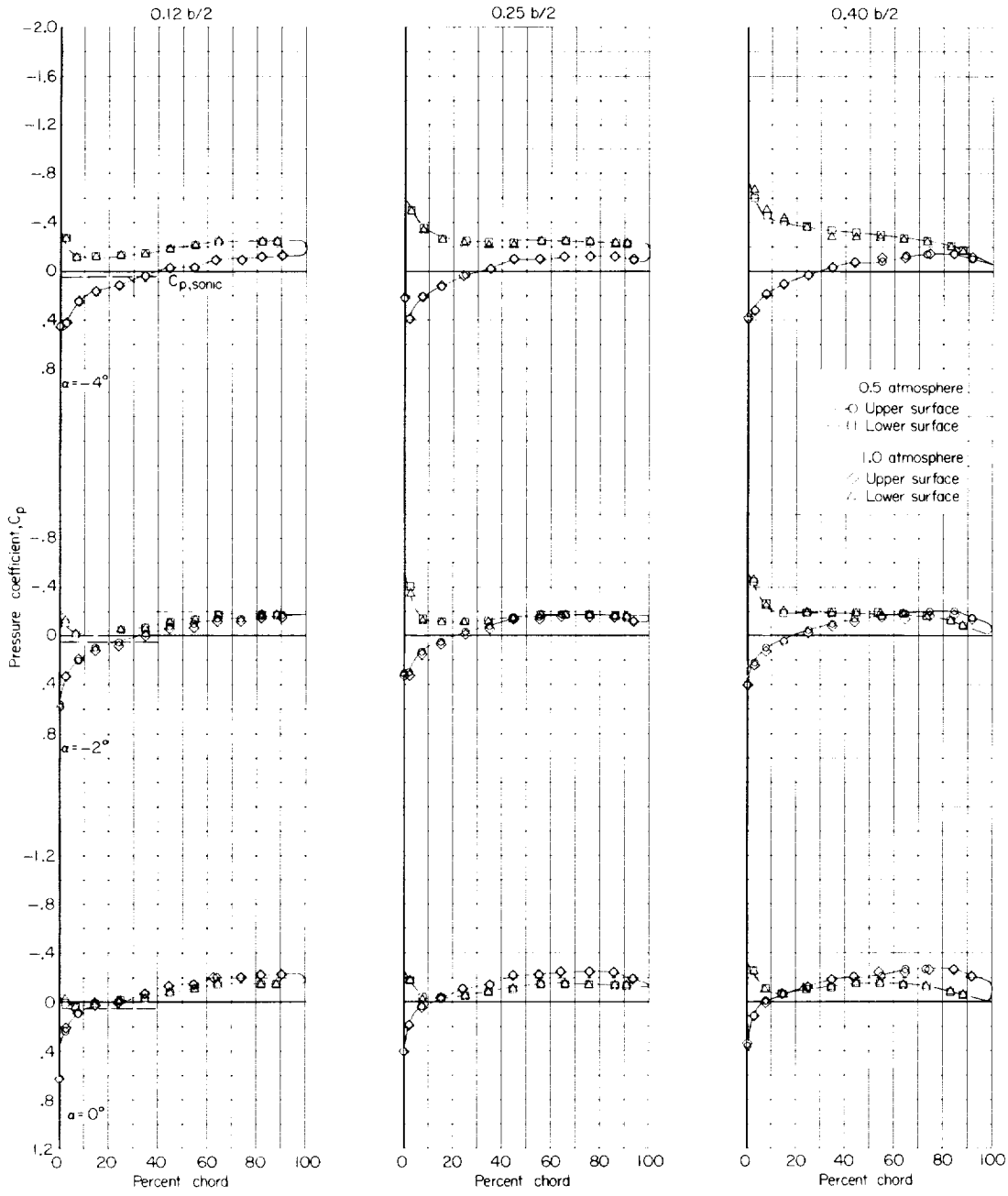
(l) M = 0.980; α = 12°, 16°, and 20°.

Figure 4.- Continued.



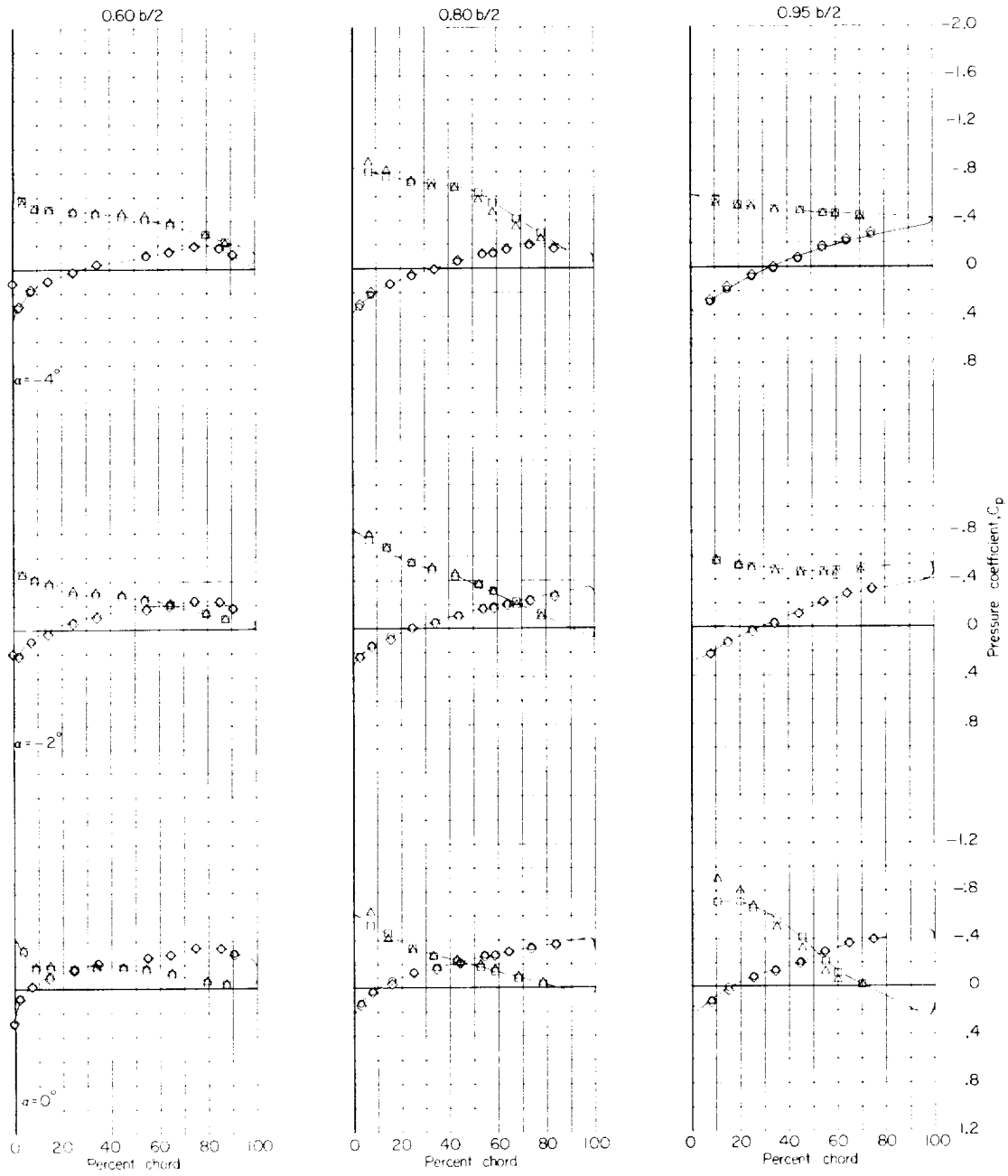
(l) Concluded.

Figure 4.- Continued.



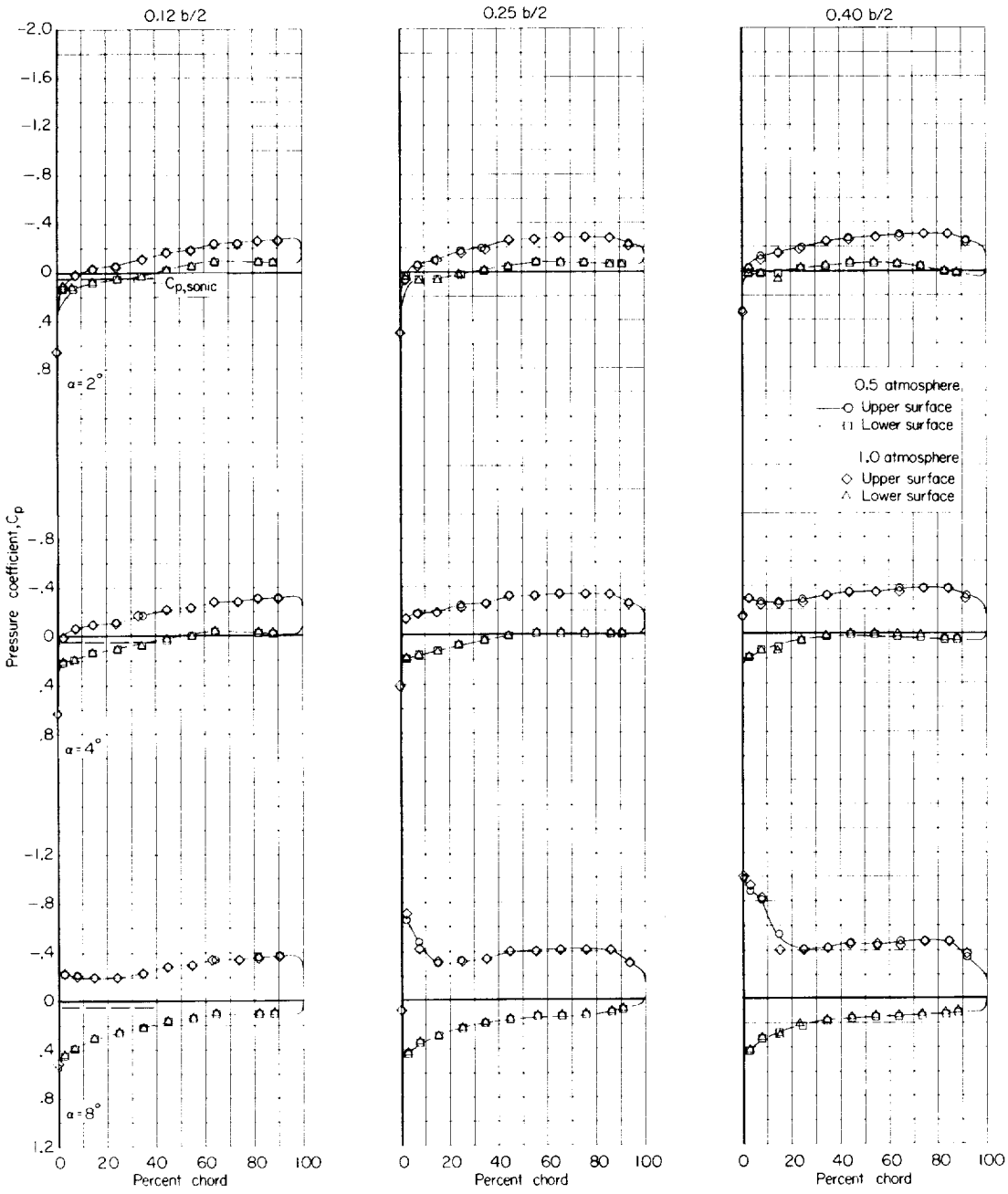
(m) M = 1.030; α = -4°, -2°, and 0°.

Figure 4.- Continued.



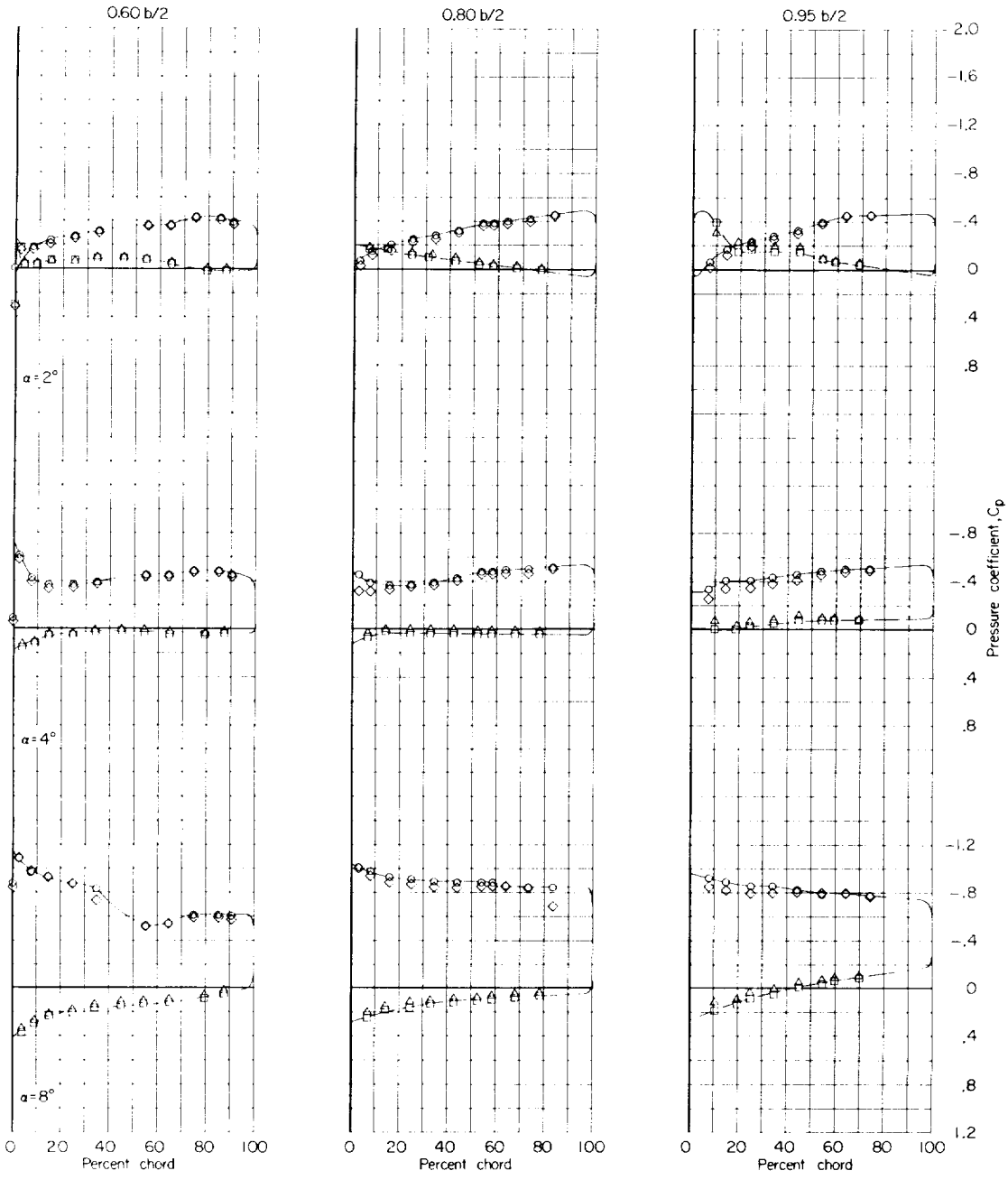
(m) Concluded.

Figure 4.- Continued.



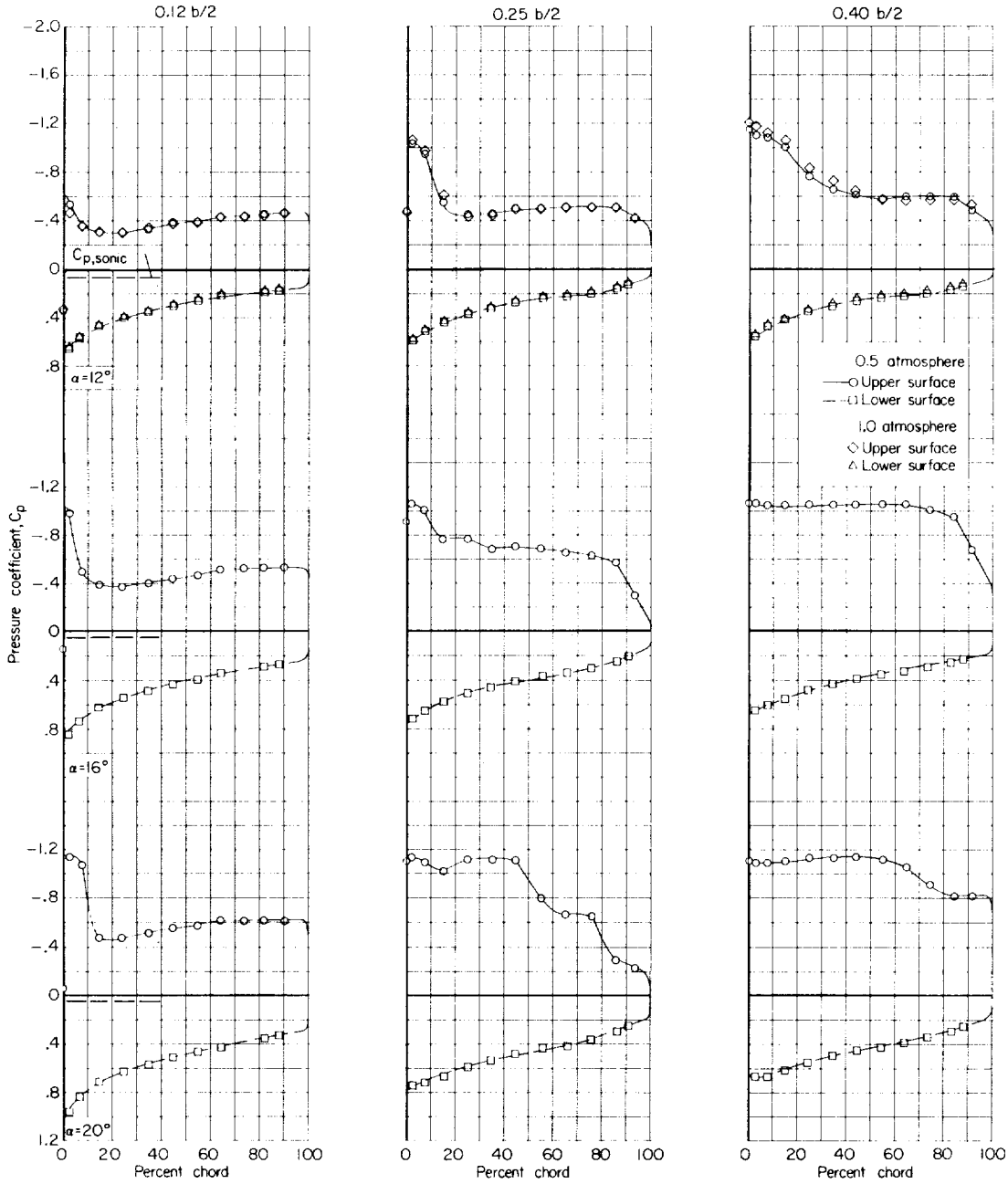
(n) M = 1.030; $\alpha = 2^{\circ}$, 4°, and 8°.

Figure 4.- Continued.



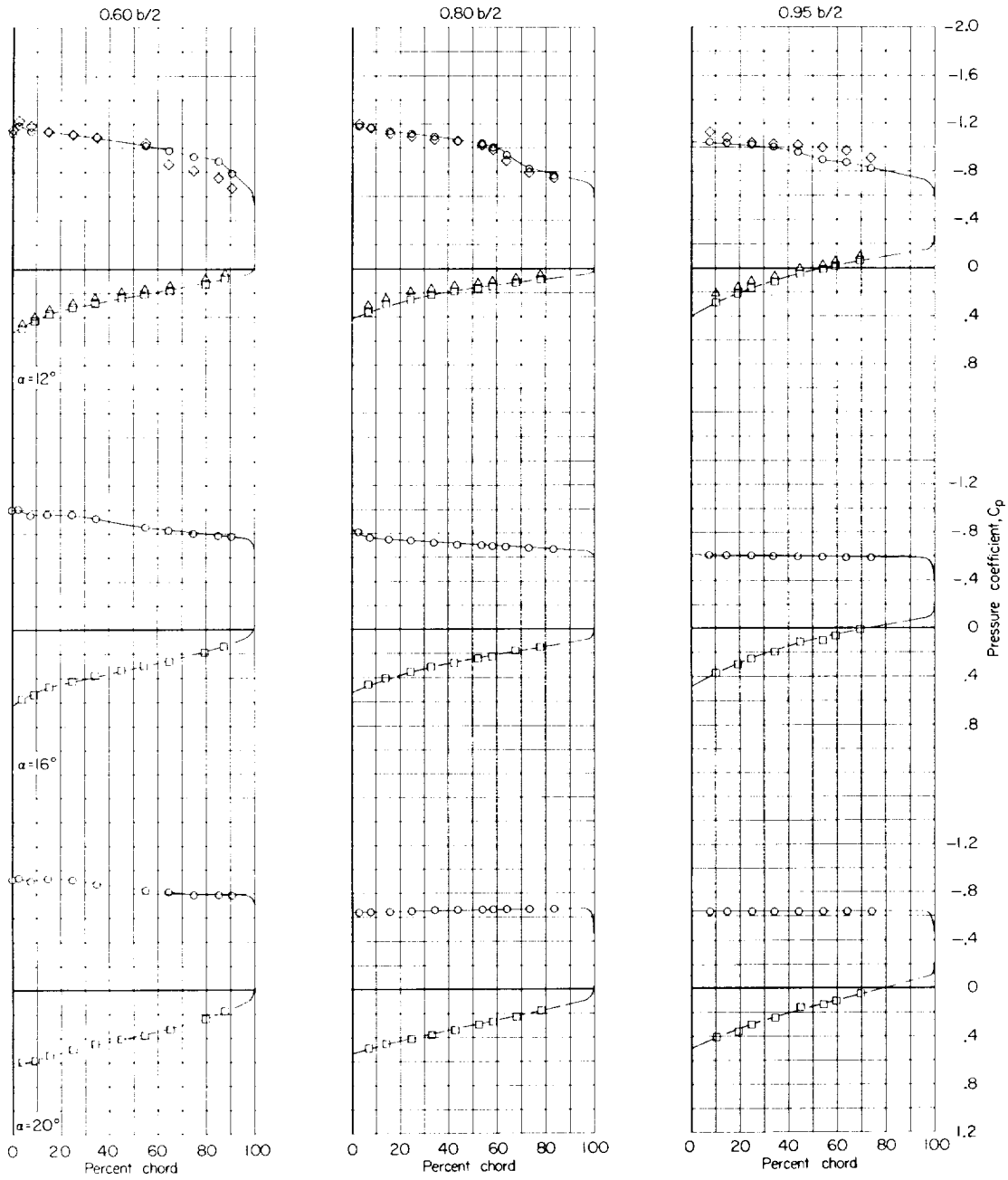
(n) Concluded.

Figure 4.- Continued.



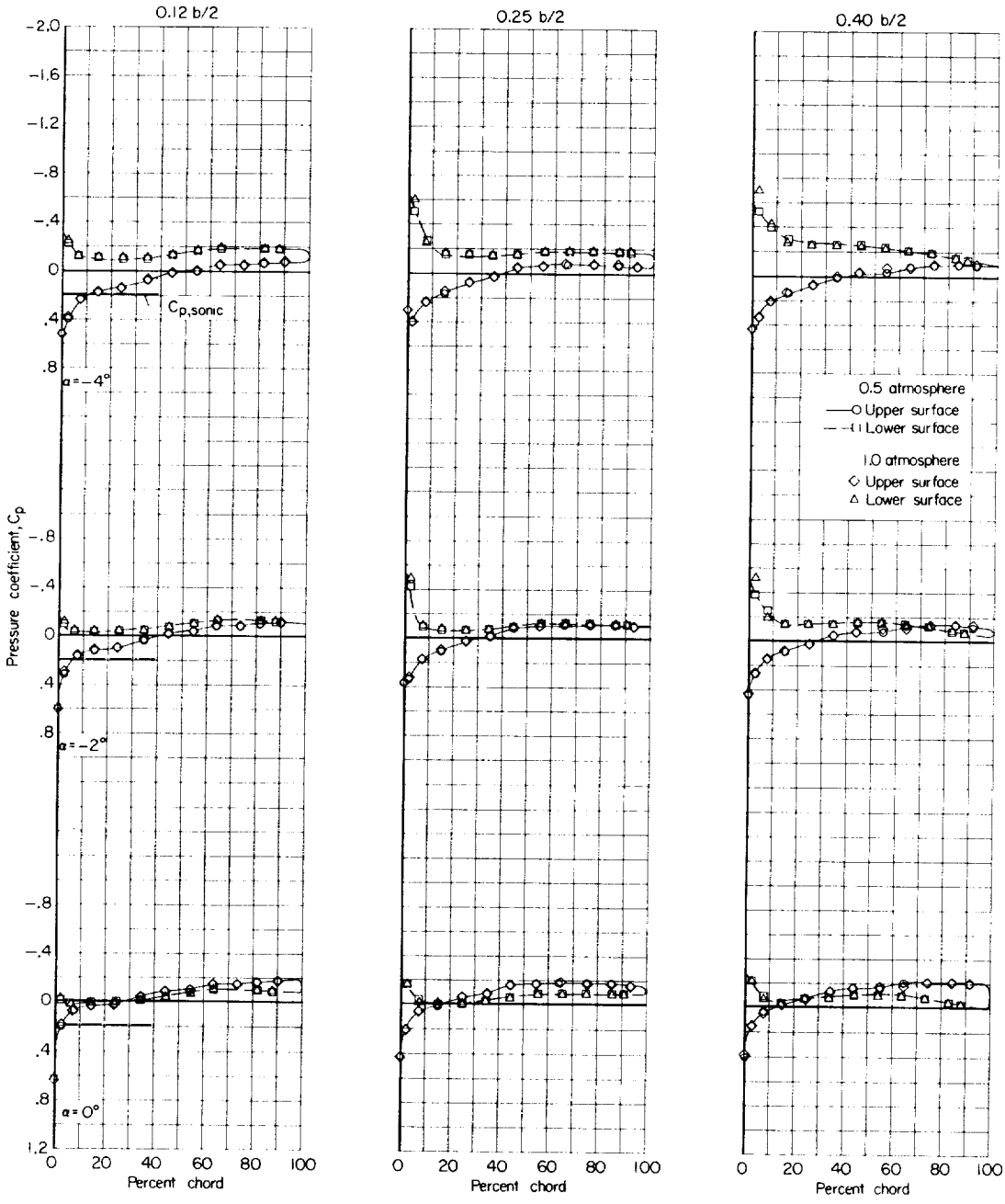
(o) M = 1.030; α = 12°, 16°, and 20°.

Figure 4.- Continued.



(o) Concluded.

Figure 4.- Continued.



(p) M = 1.125; α = -4°, -2°, and 0°.

Figure 4.- Continued.

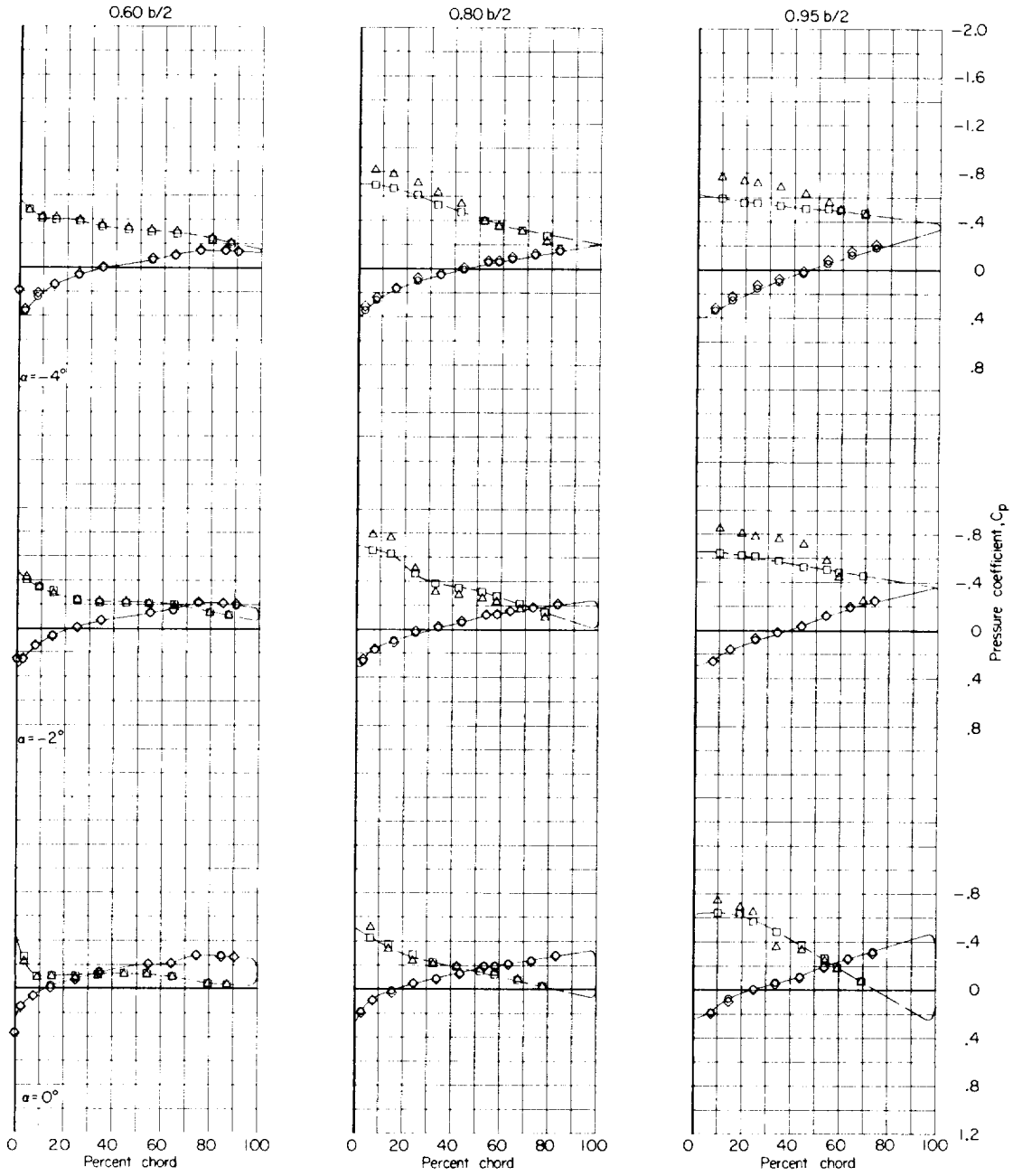


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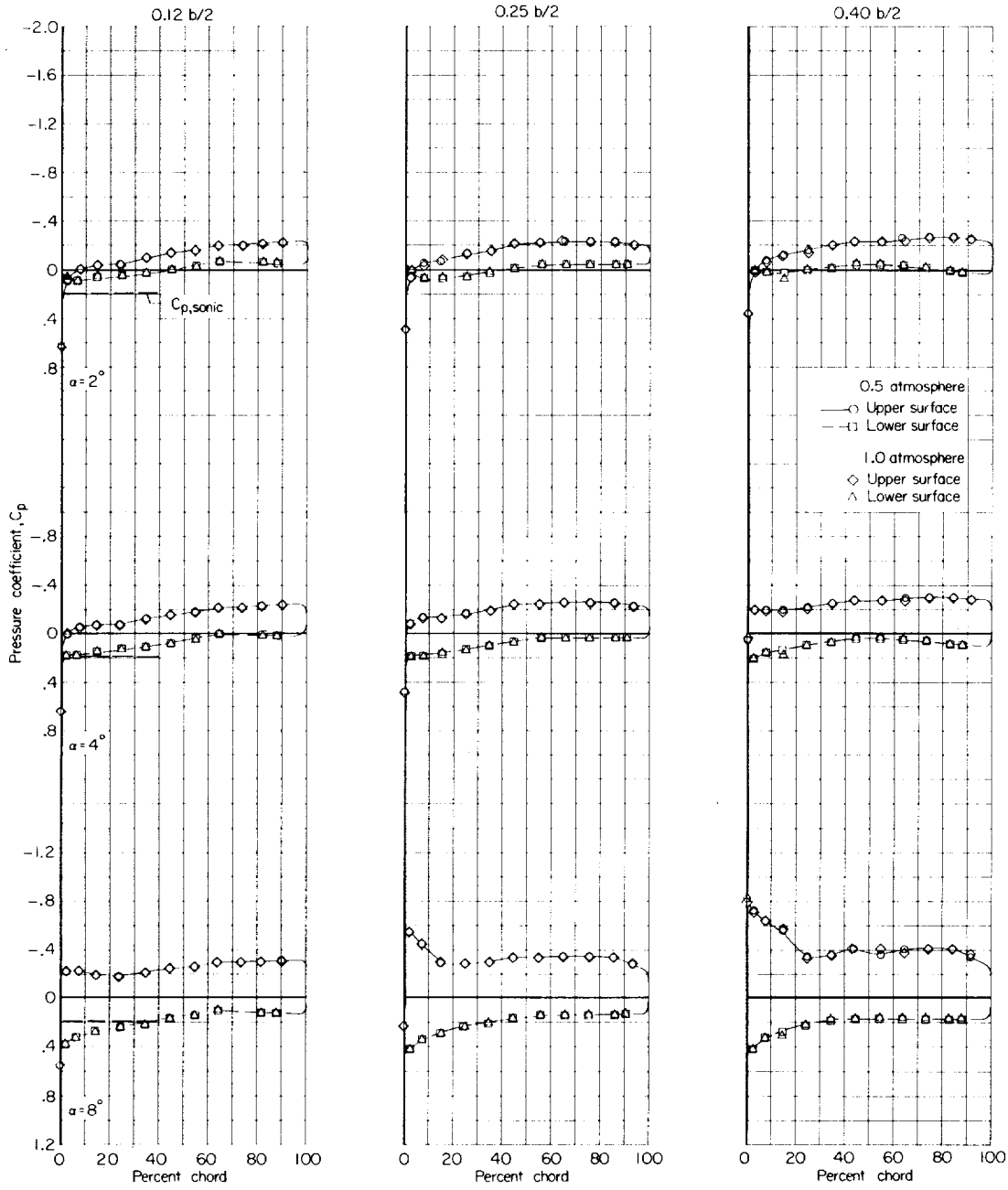
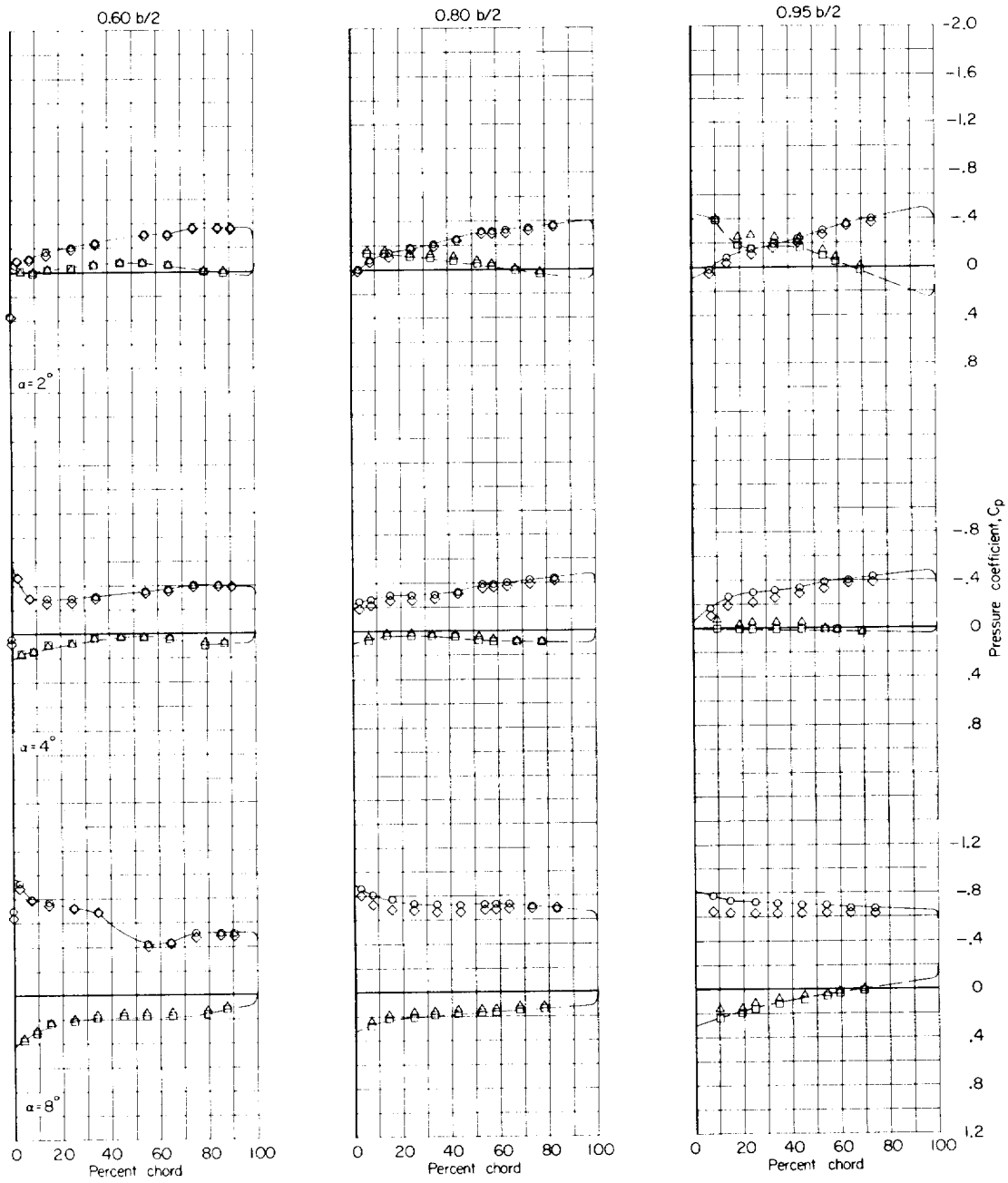
Figure 4.- Continued.



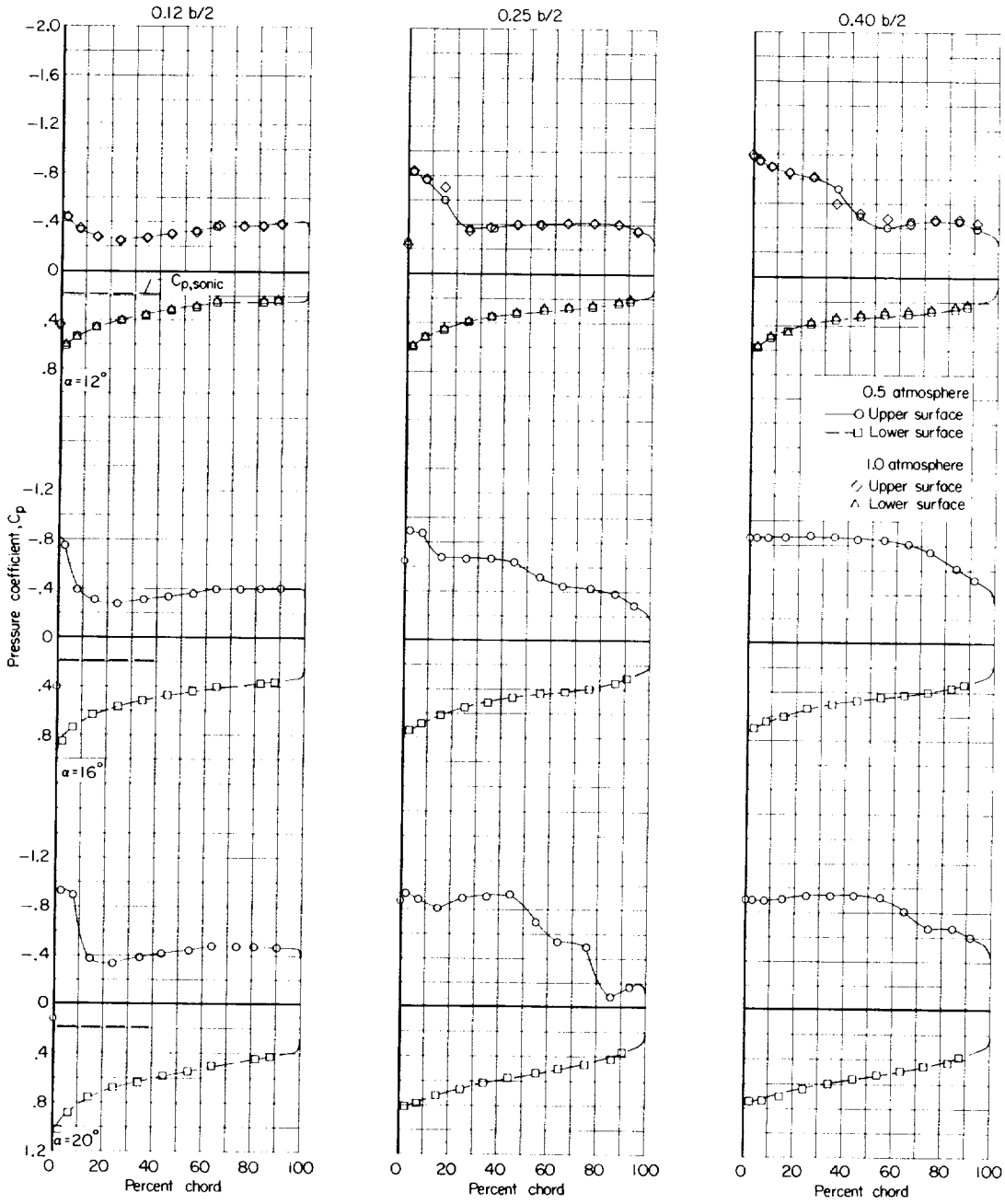
(q) M = 1.125; α = 2°, 4°, and 8°.

Figure 4.- Continued.



(q) Concluded.

Figure 4.- Continued.



(r) M = 1.125; α = 12°, 16°, and 20°.

Figure 4.- Continued.

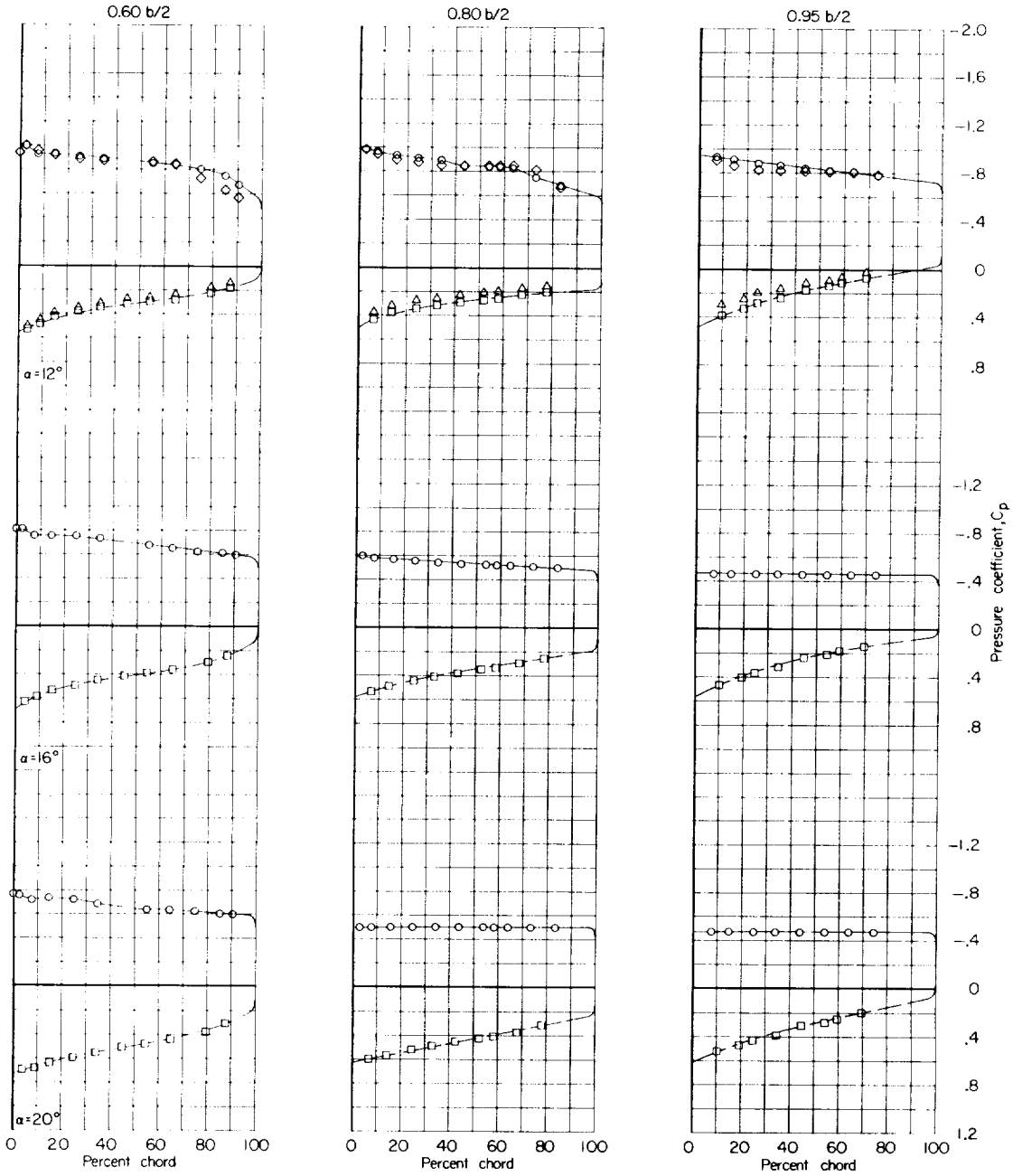


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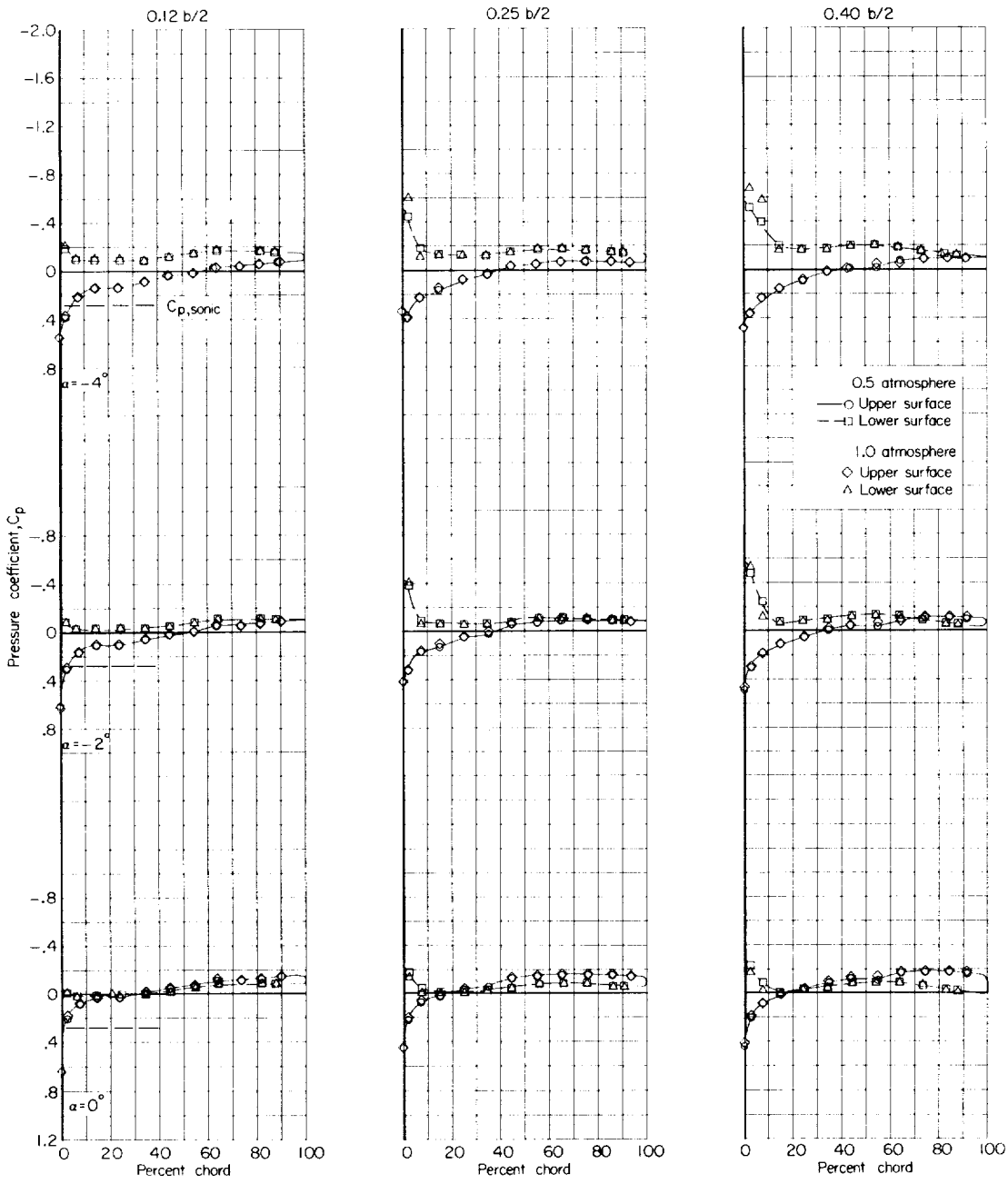
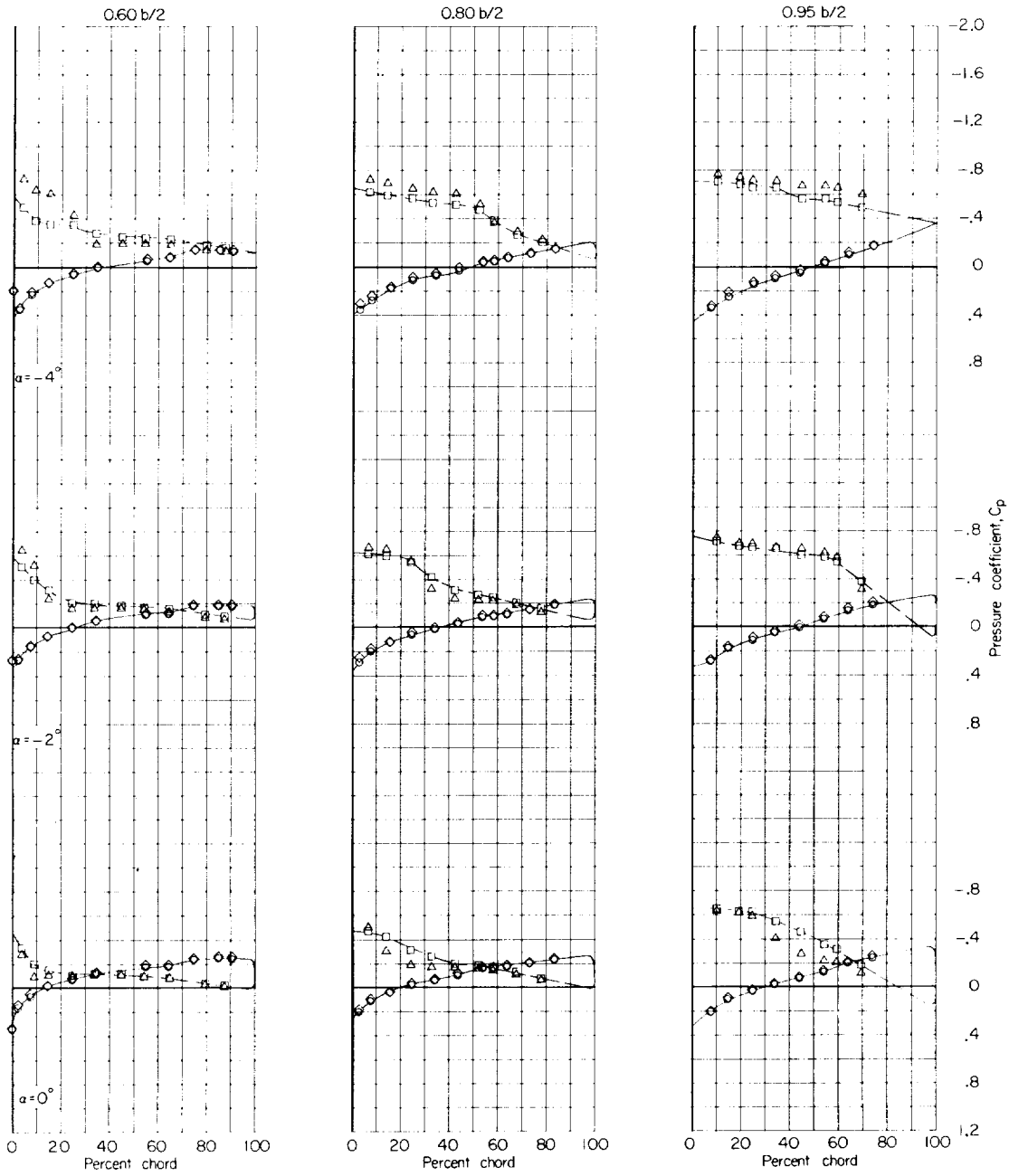
Figure 4.- Continued.



(s) M = 1.200; α = -4°, -2°, and 0°.

Figure 4.- Continued.



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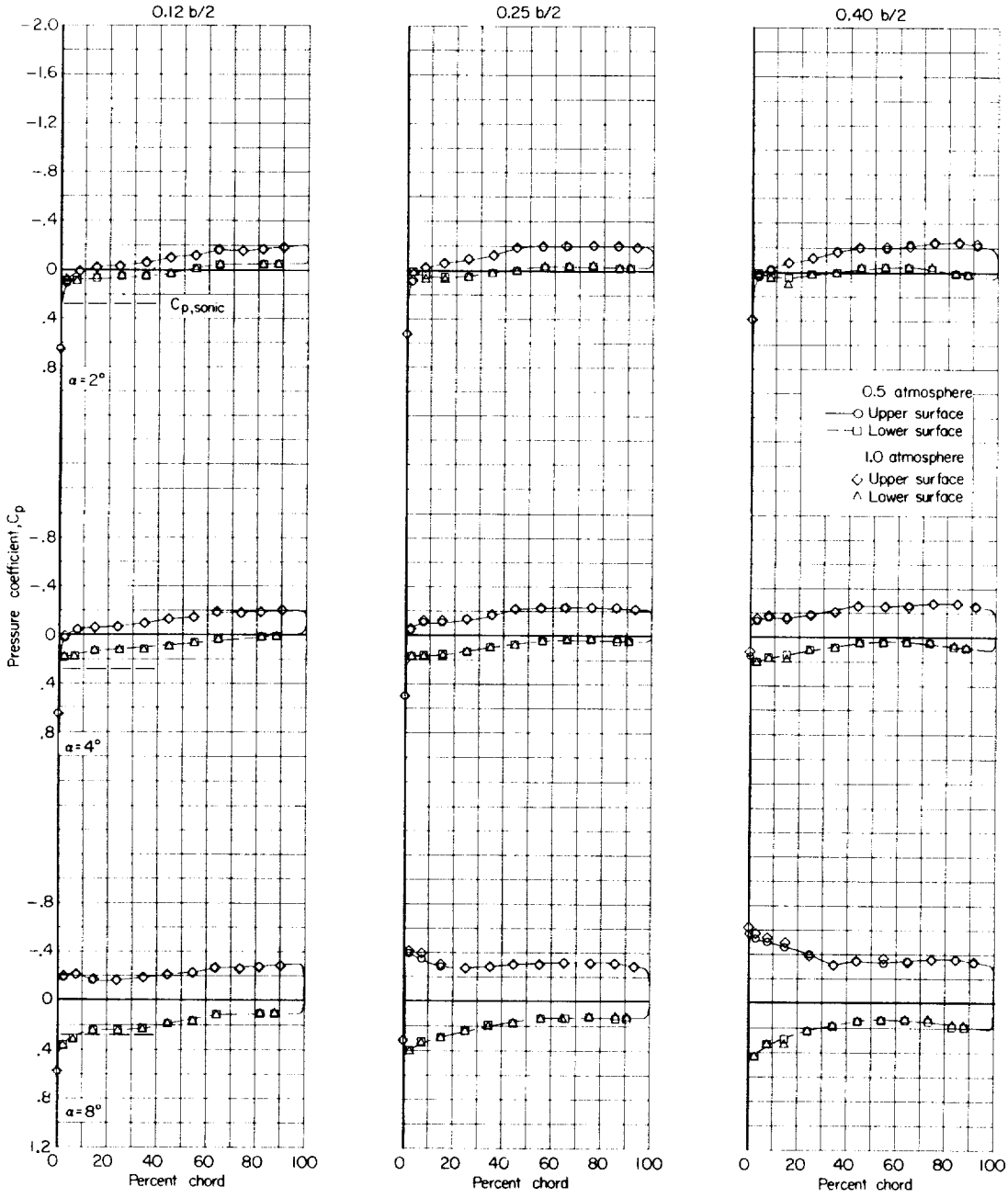
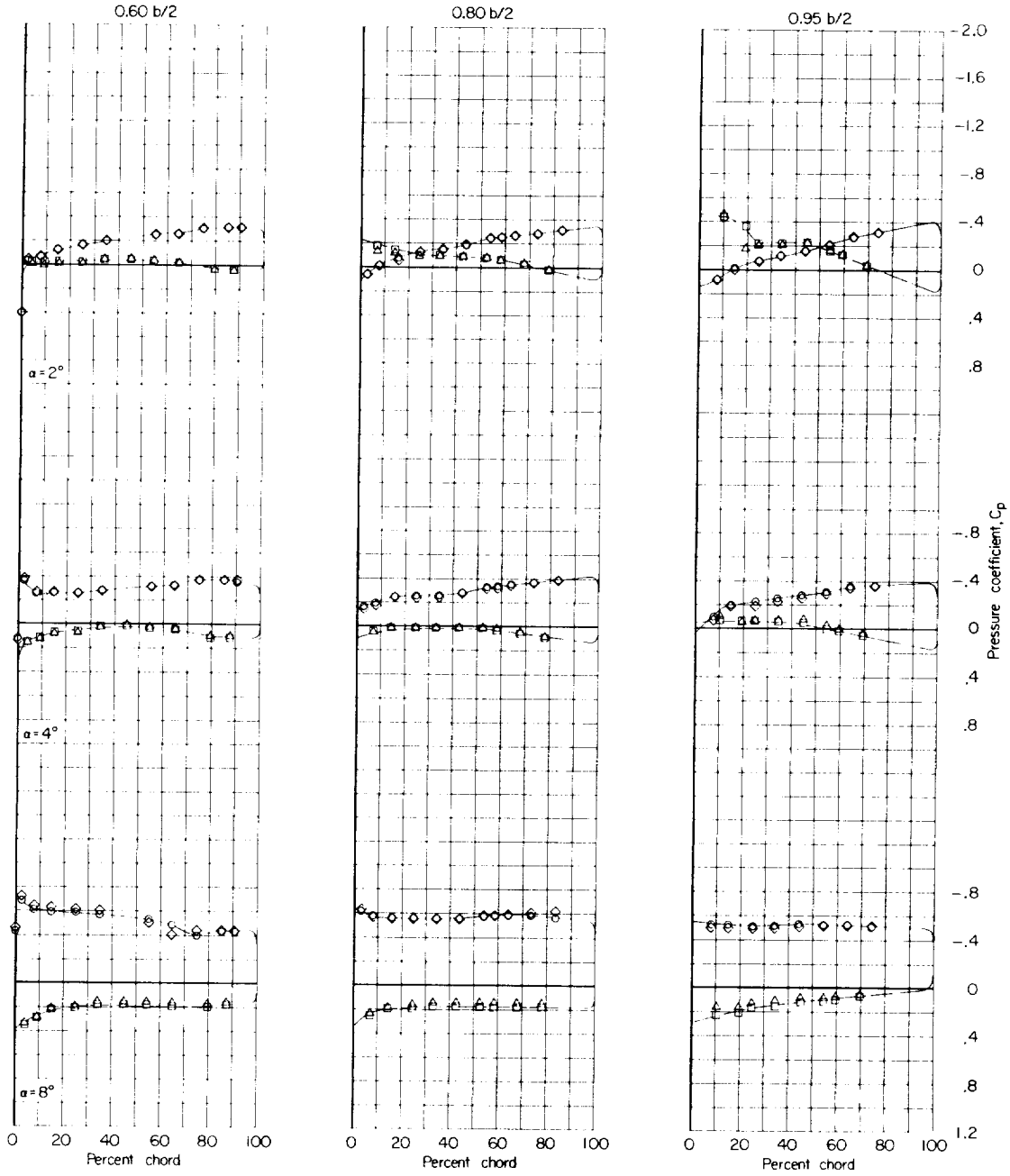


(t) M = 1.200; $\alpha = 2^{o}$, 4^{o}, and 8^{o}.

Figure 4.- Continued.



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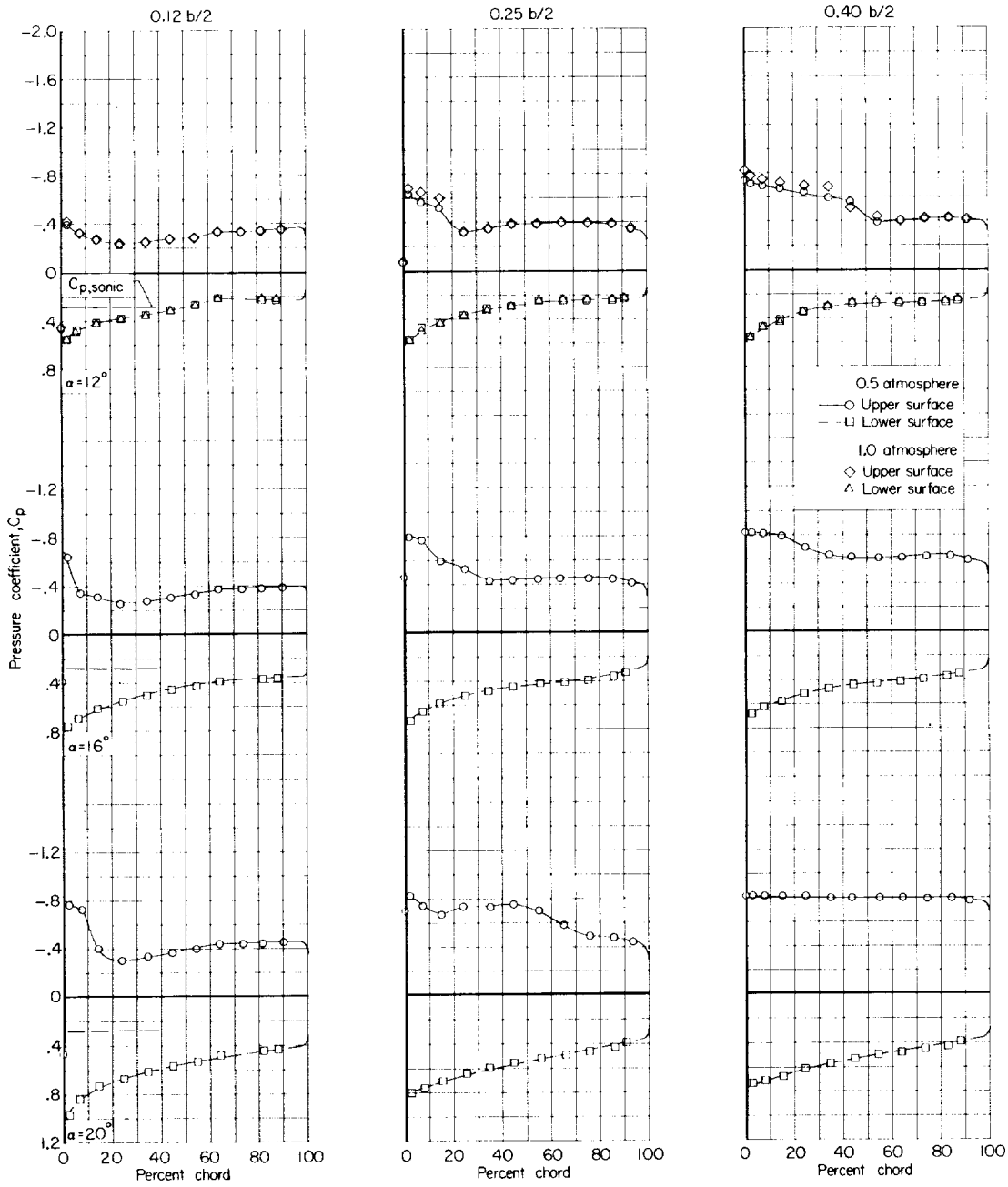
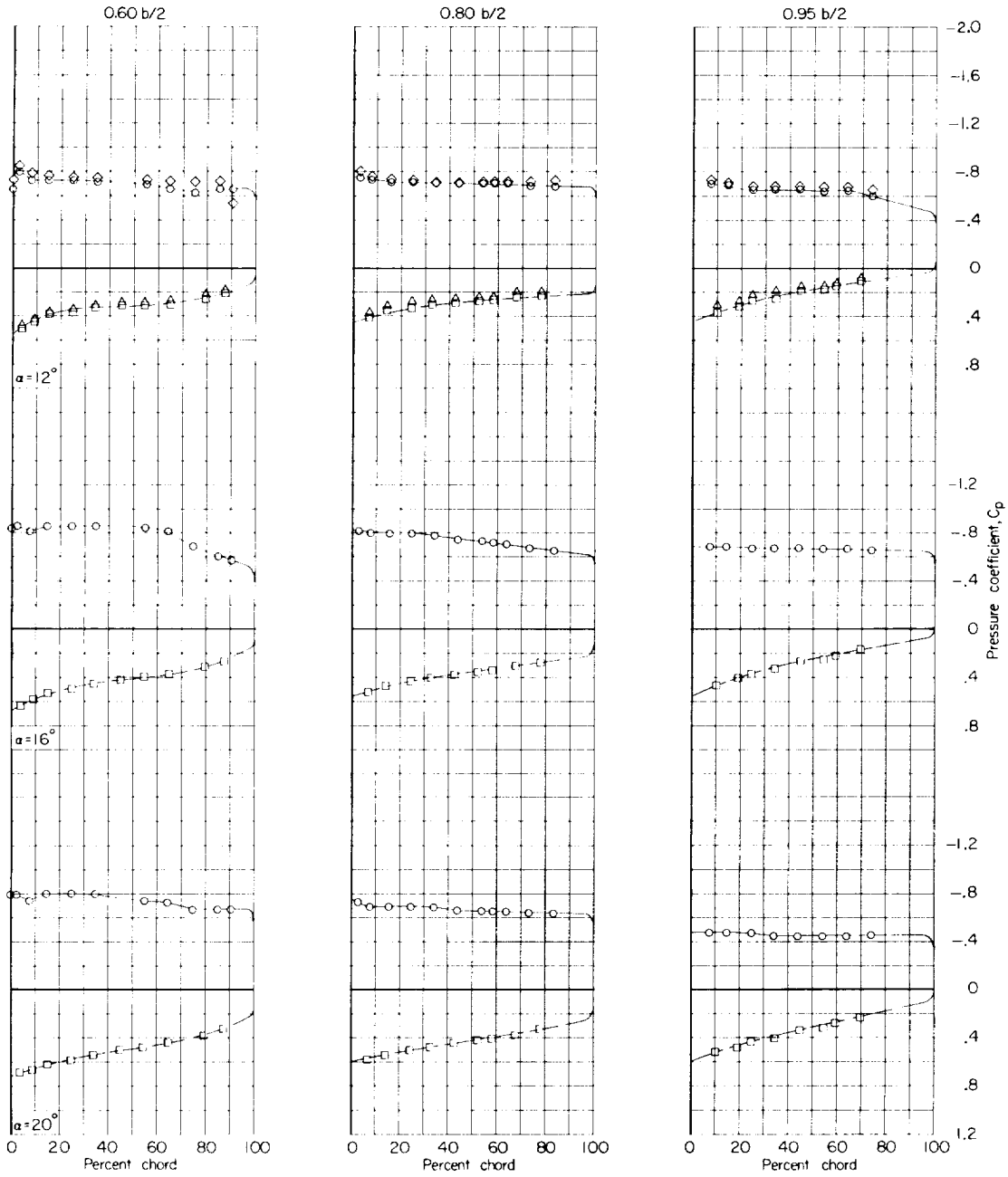


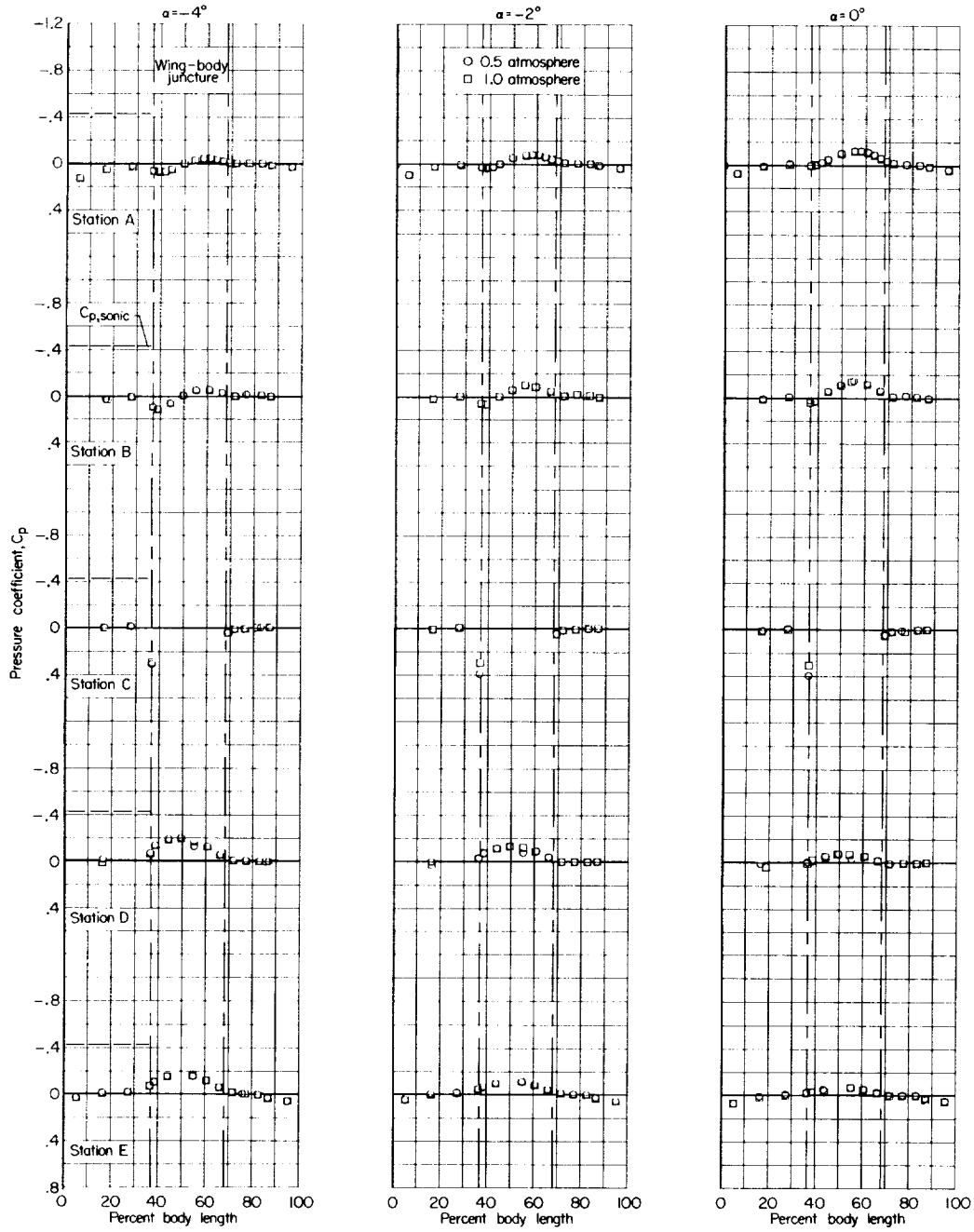
(u) M = 1.200; α = 12°, 16°, and 20°.

Figure 4.- Continued.



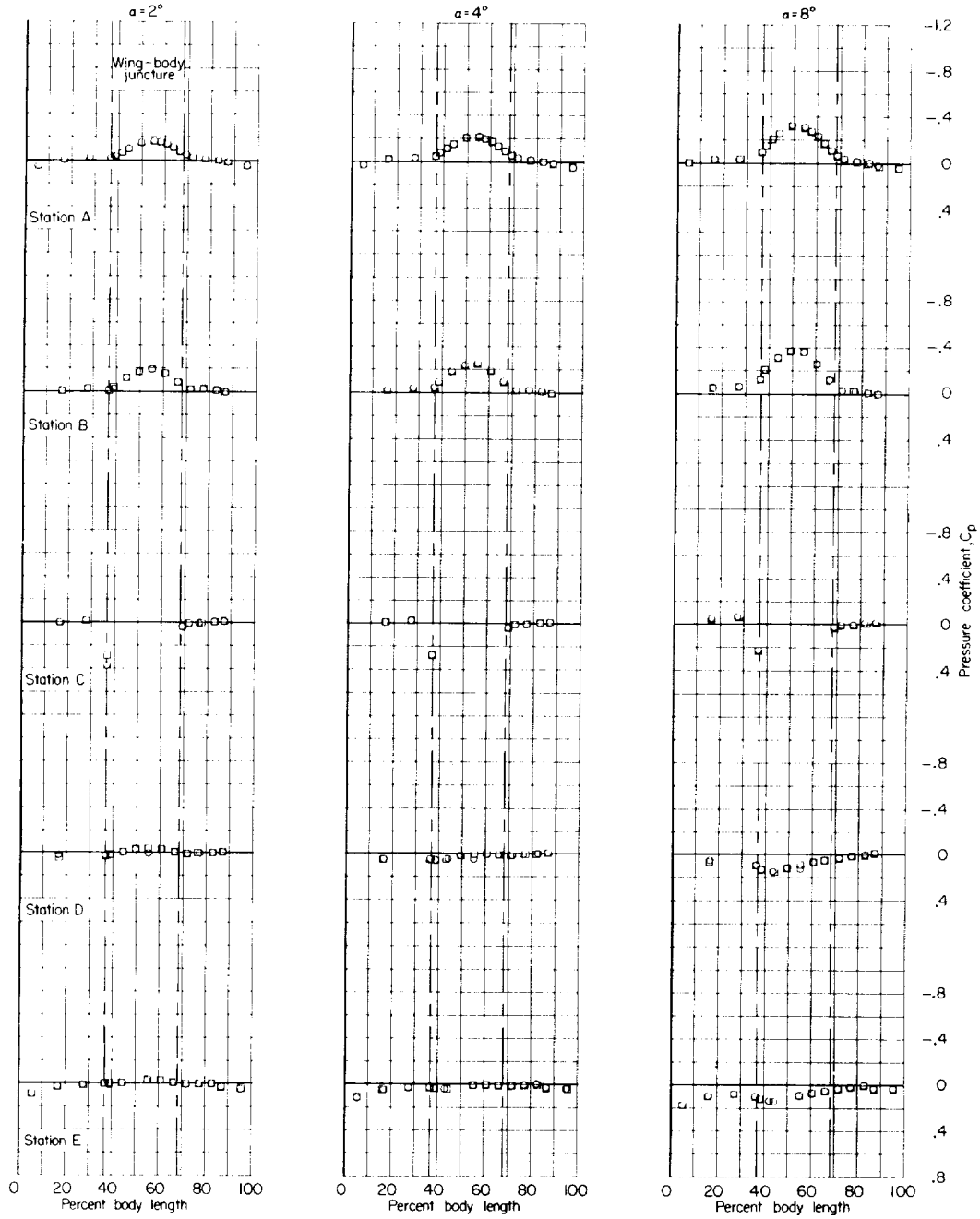
(u) Concluded.

Figure 4.- Concluded.



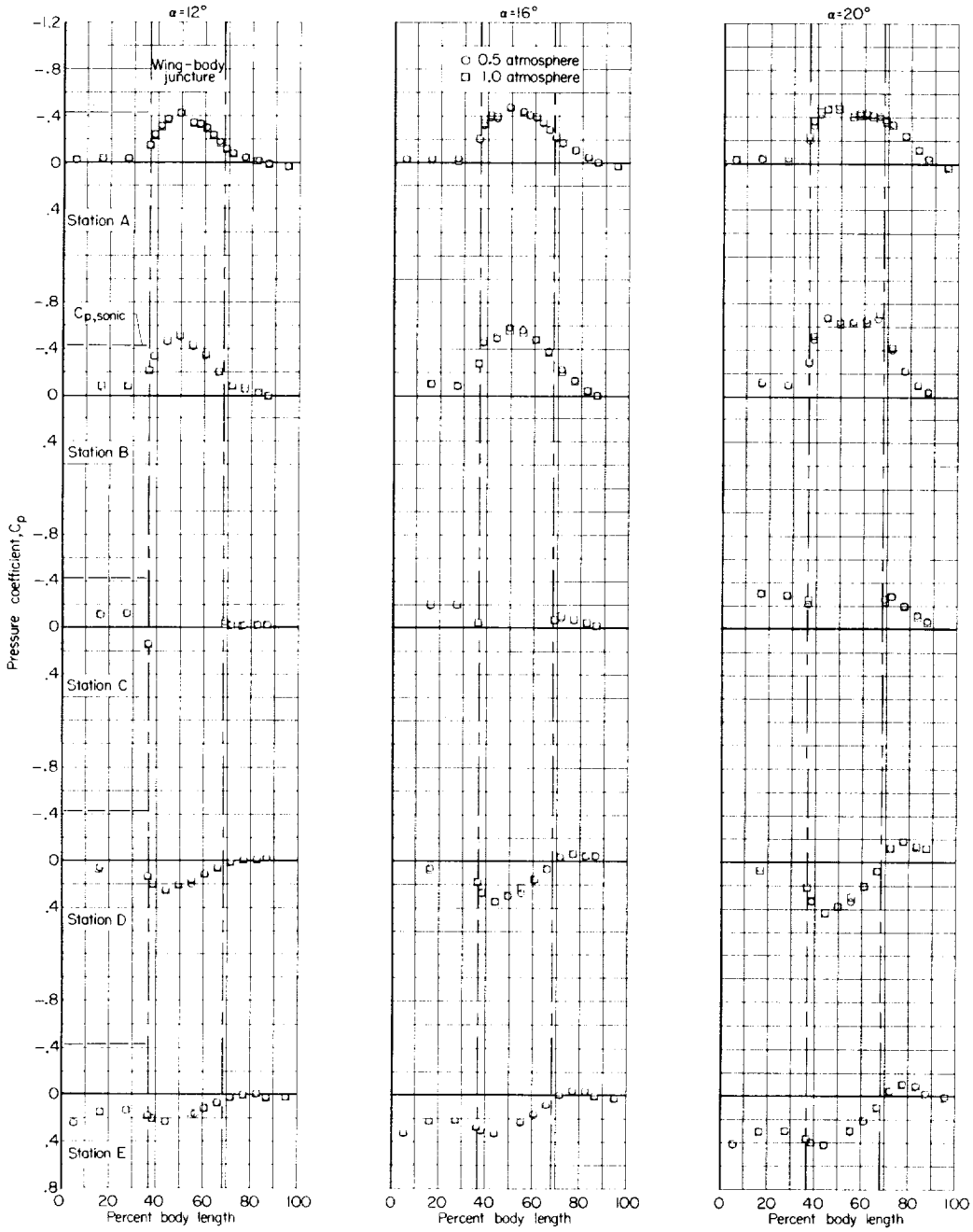
(a) M = 0.800.

Figure 5.- Pressure measurements on the body in the presence of the wing.



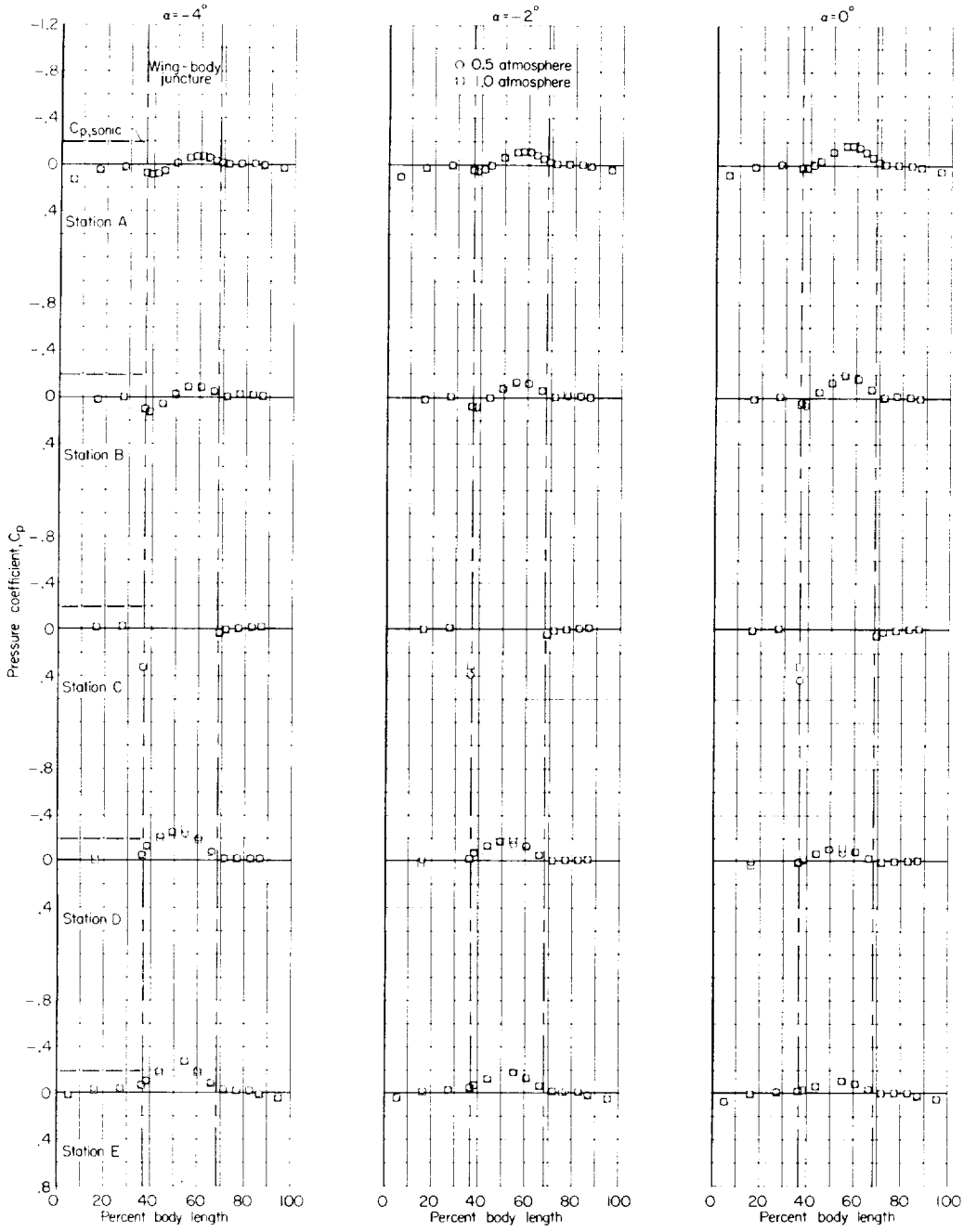
(a) Continued.

Figure 5.- Continued.



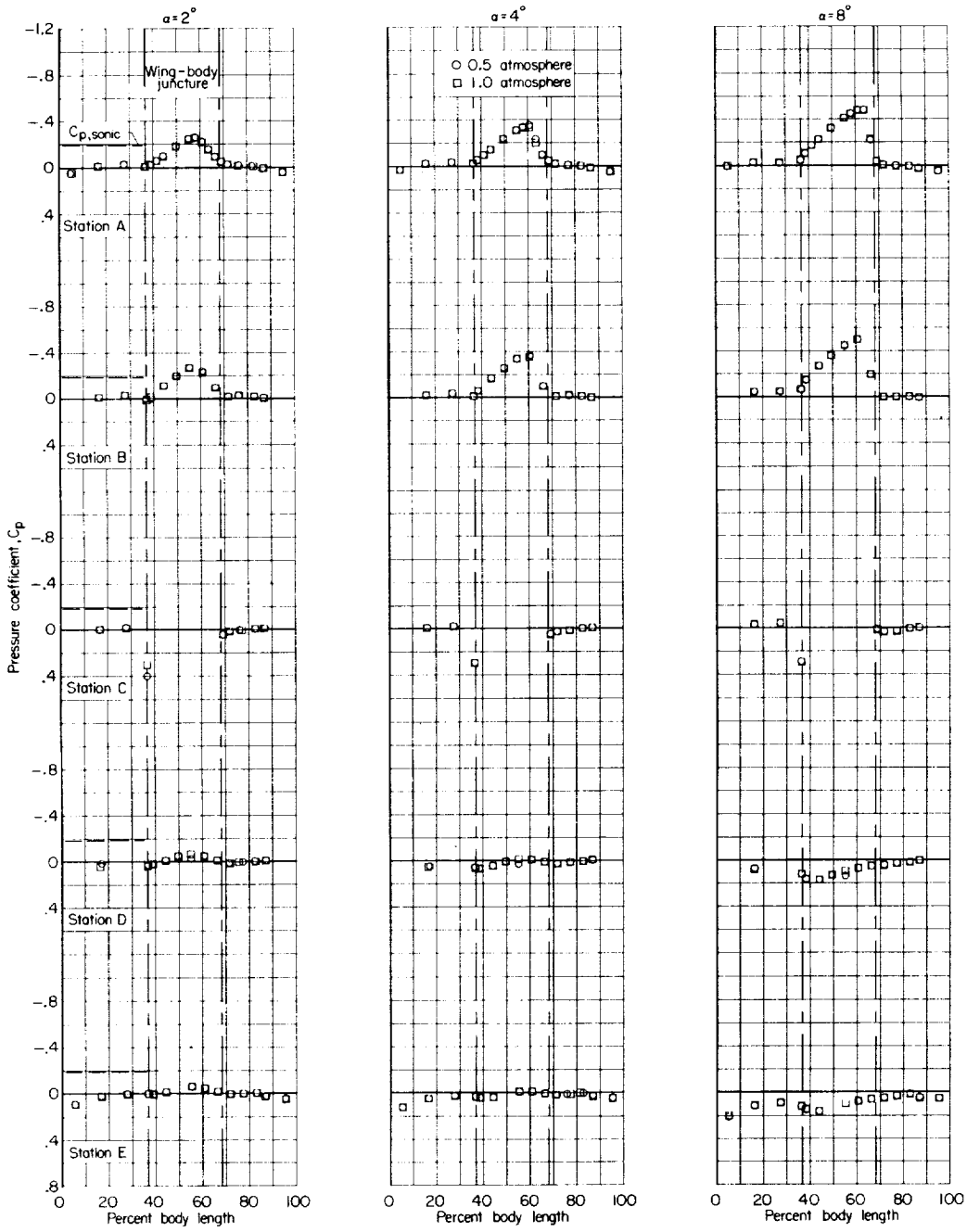
(a) Concluded.

Figure 5.- Continued.



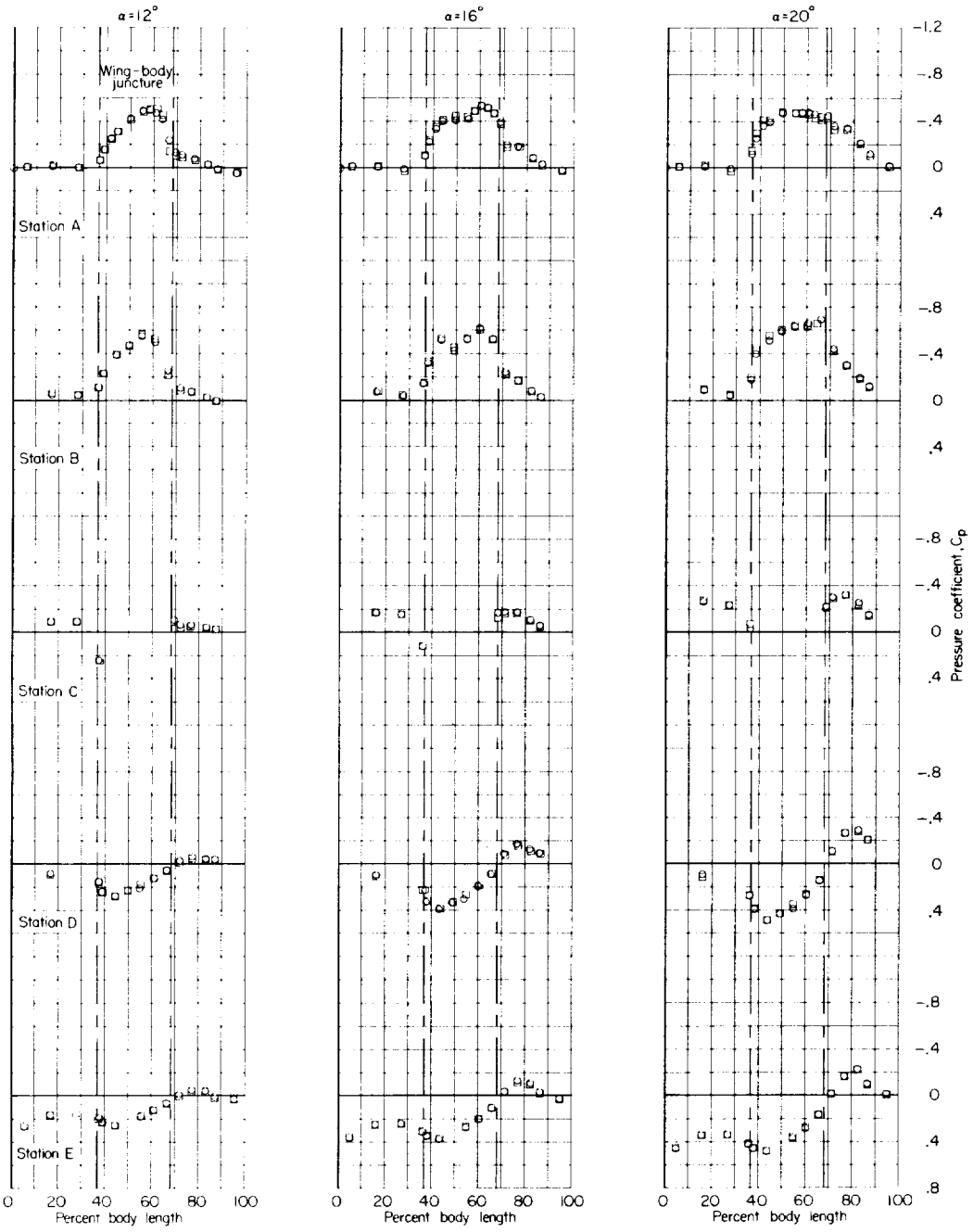
(b) M = 0.900.

Figure 5.- Continued.



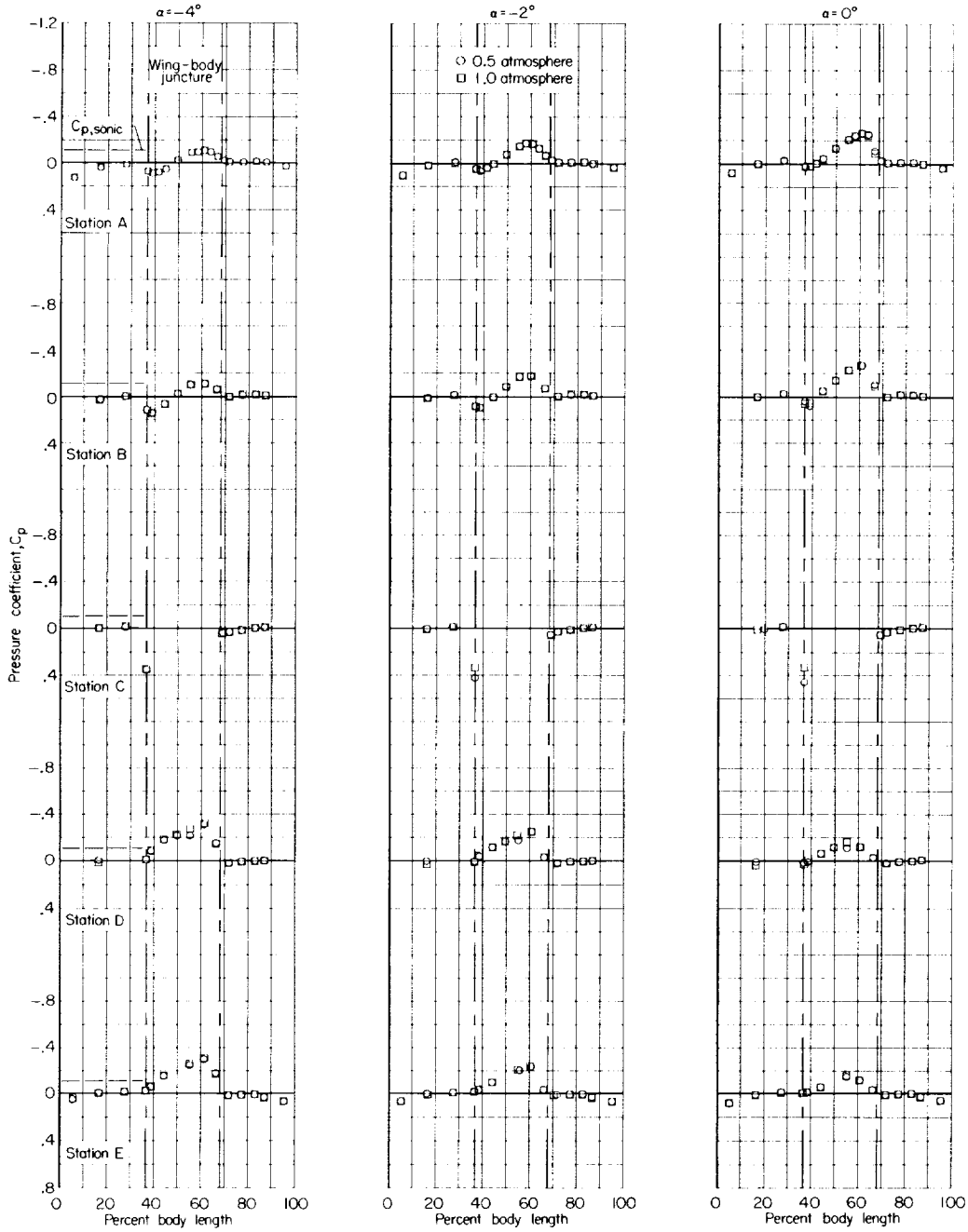
(b) Continued.

Figure 5.- Continued.



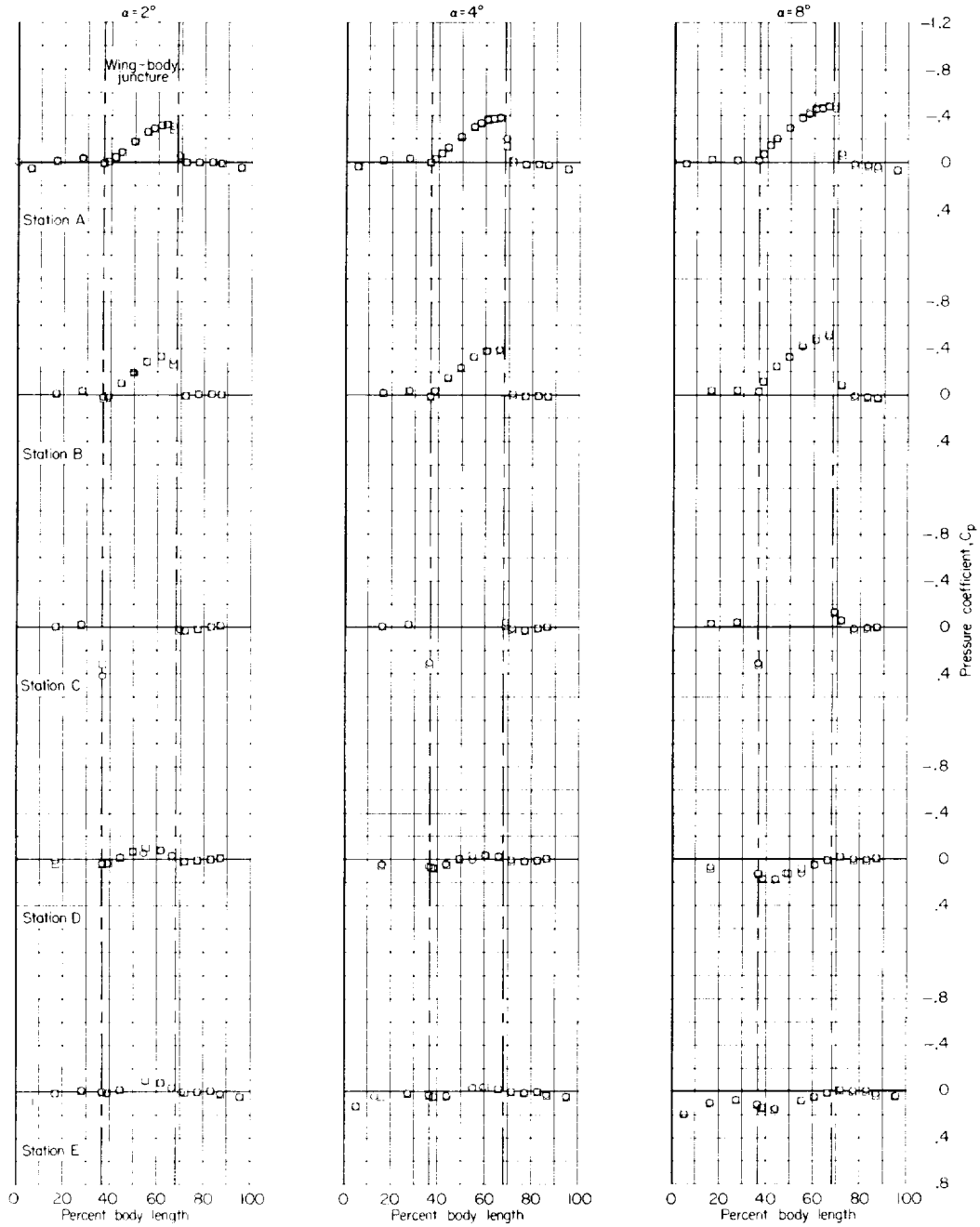
(b) Concluded.

Figure 5.- Continued.



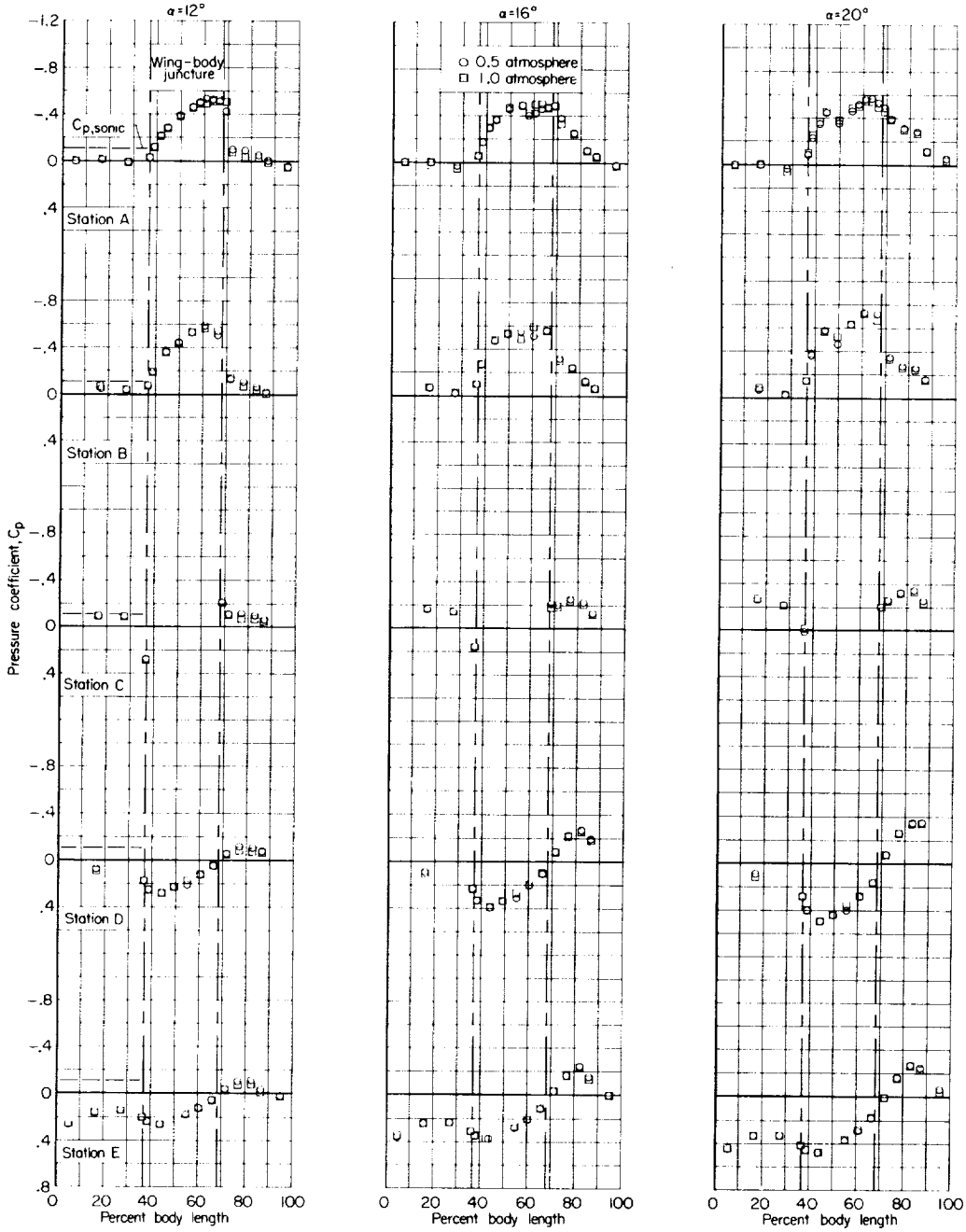
(c) M = 0.940.

Figure 5.- Continued.



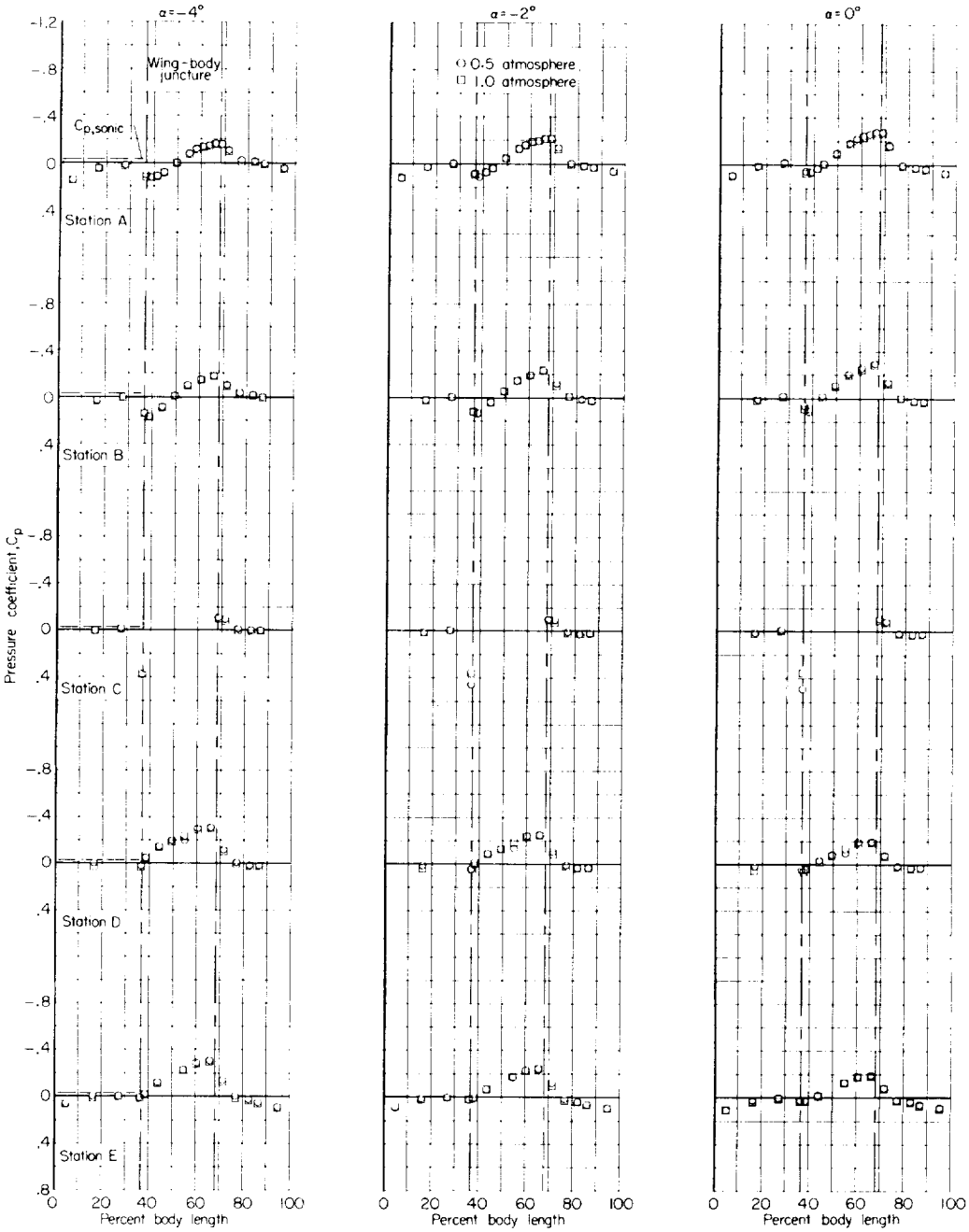
(c) Continued.

Figure 5.- Continued.



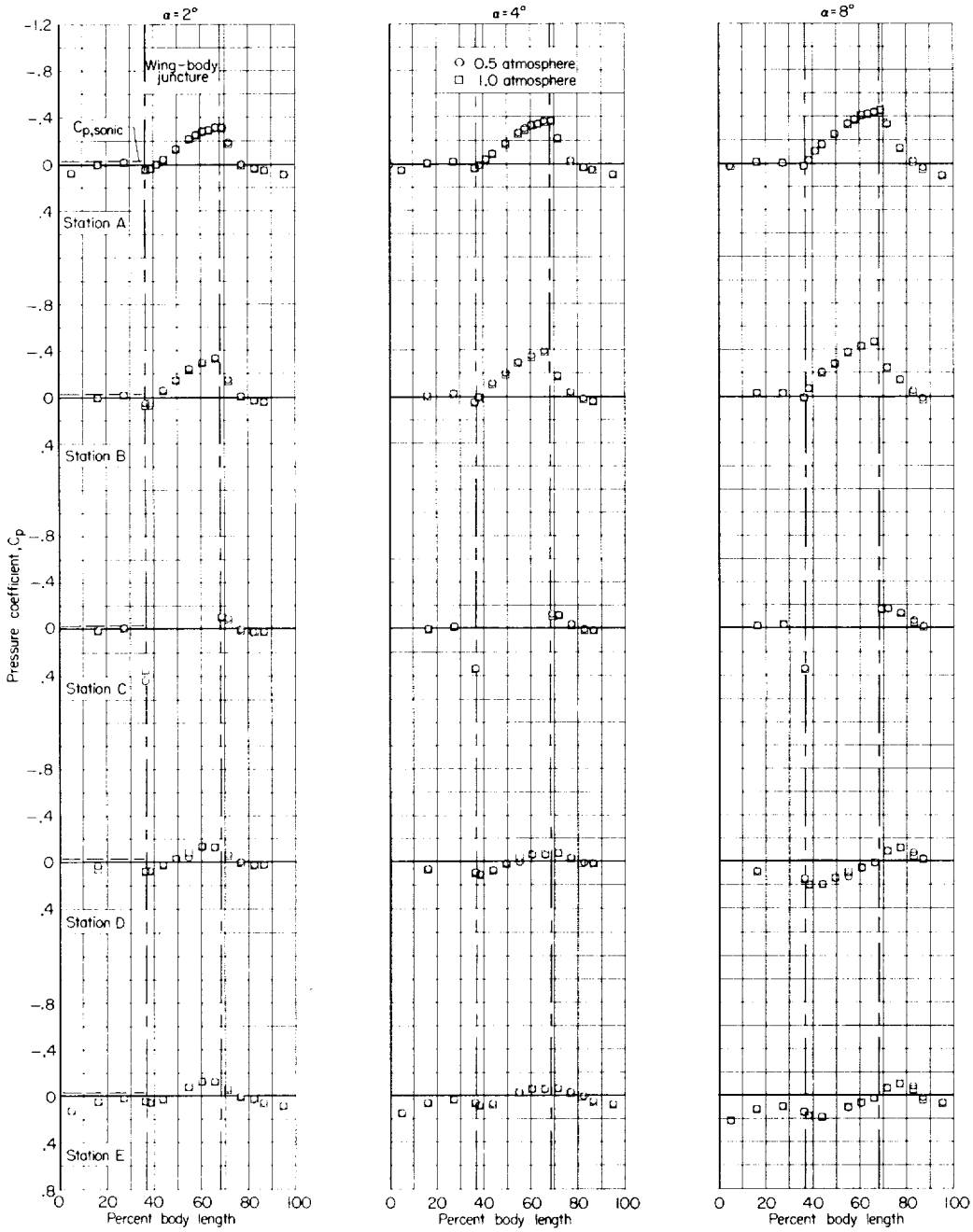
(c) Concluded.

Figure 5.- Continued.



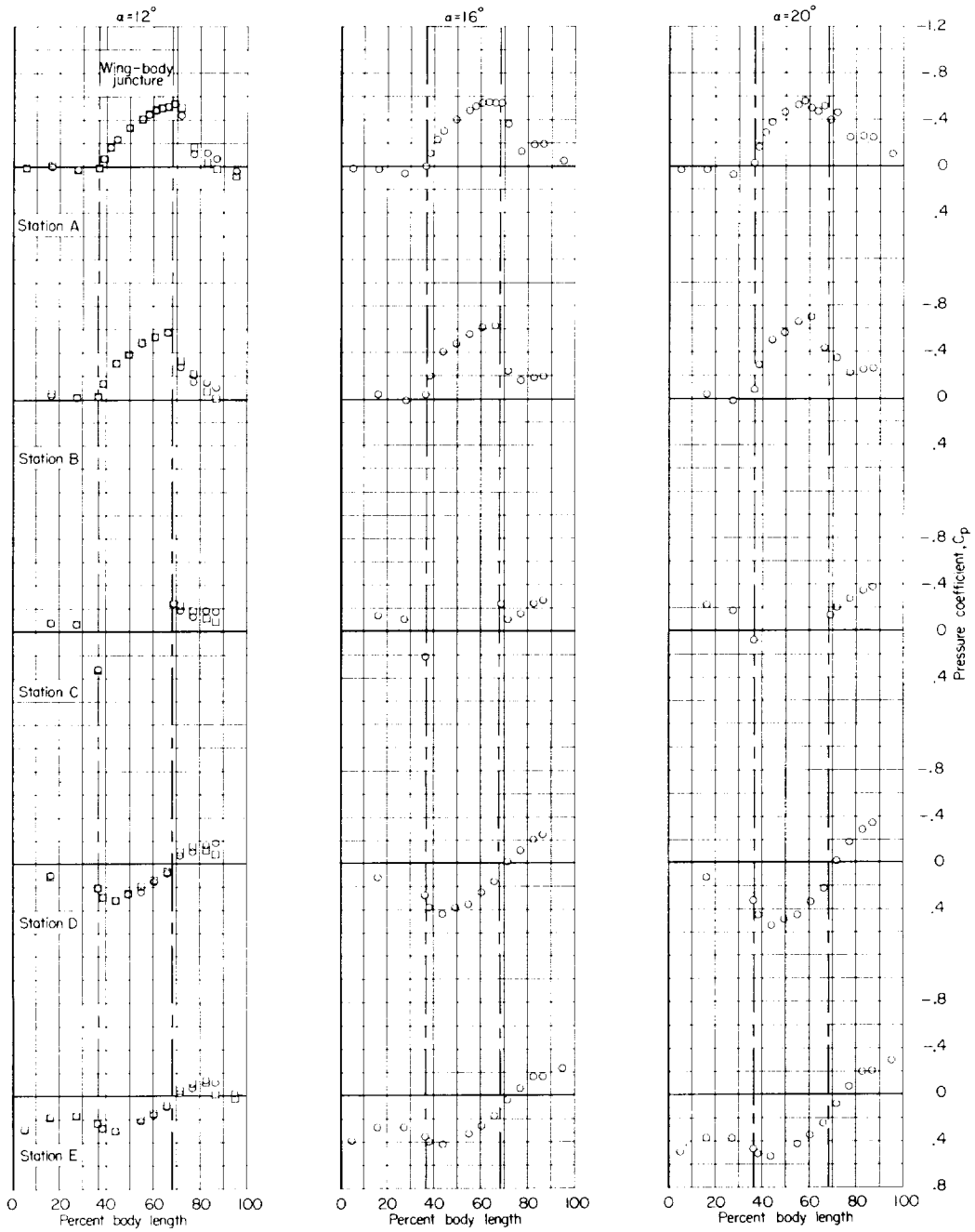
(d) M = 0.980.

Figure 5.- Continued.



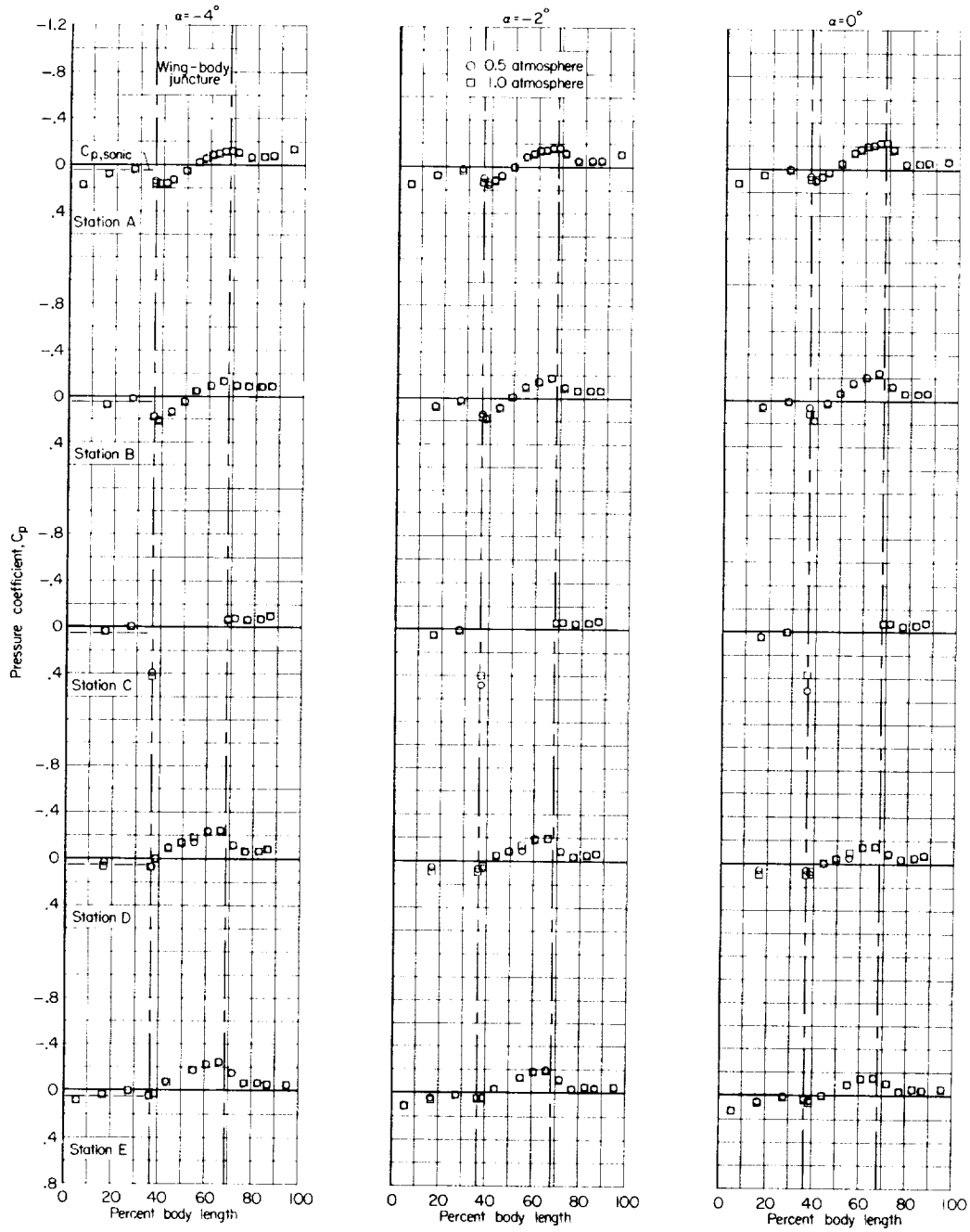
(d) Continued.

Figure 5.- Continued.



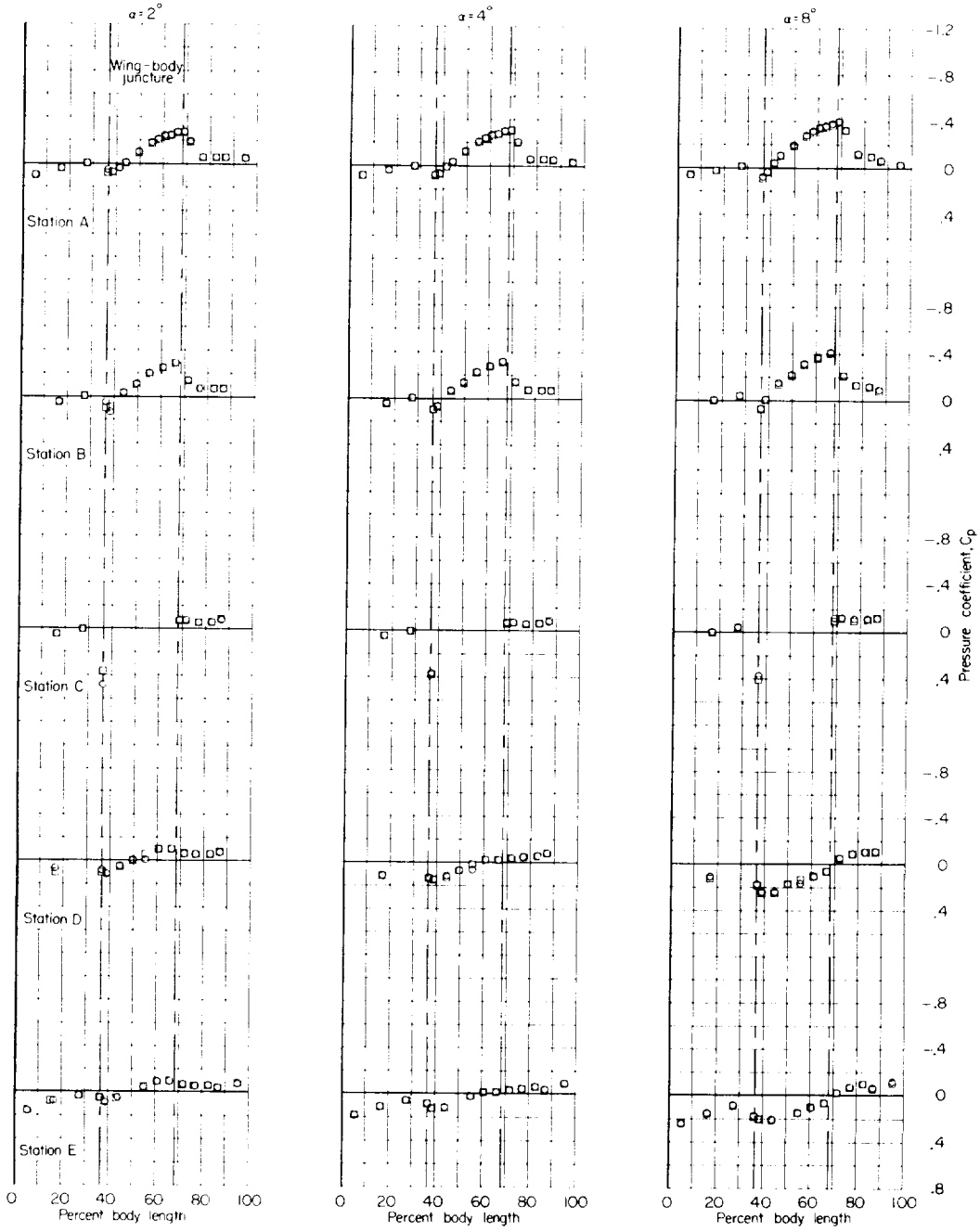
(d) Concluded.

Figure 5.- Continued.



(e) M = 1.030.

Figure 5.- Continued.



(e) Continued.

Figure 5.- Continued.

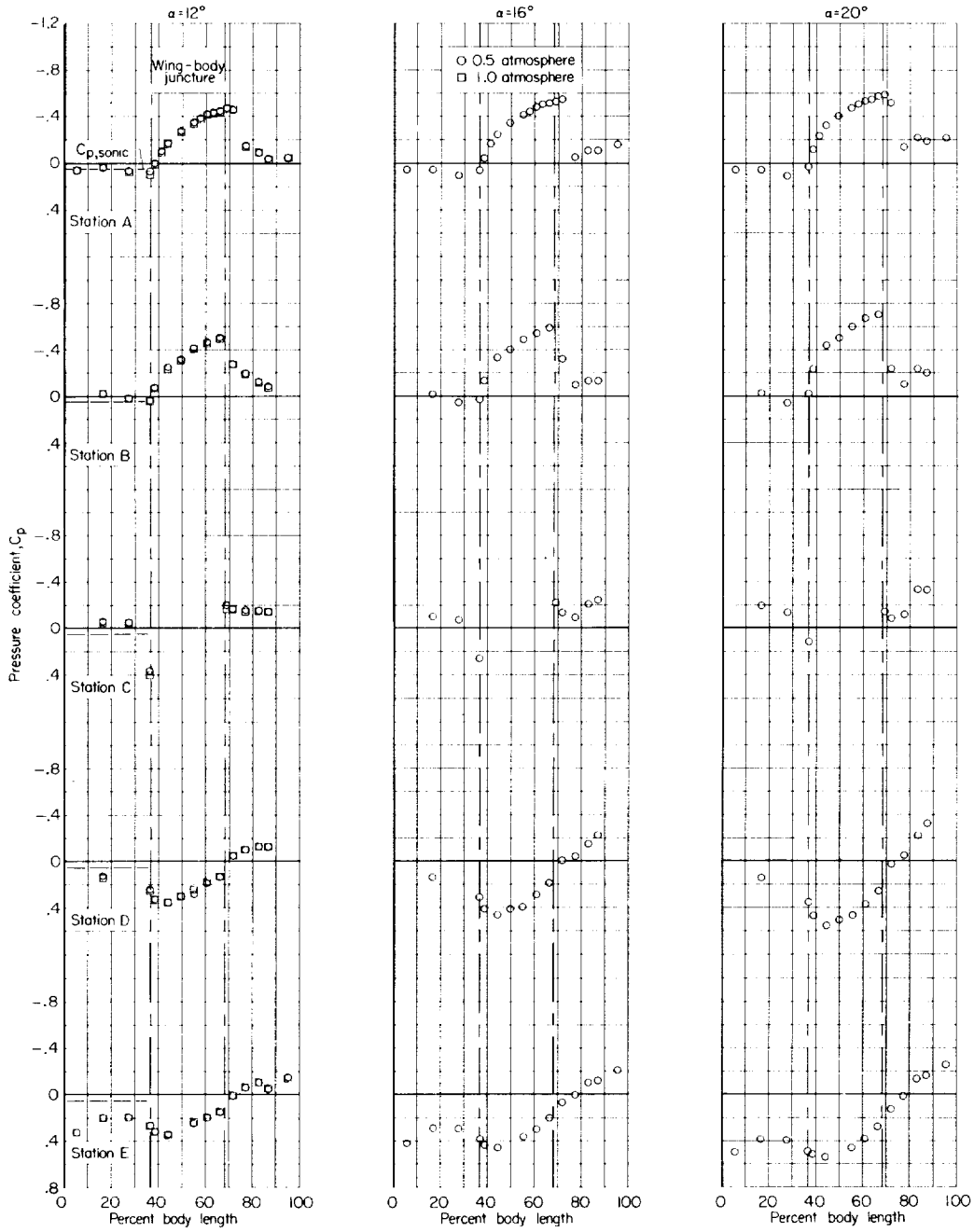


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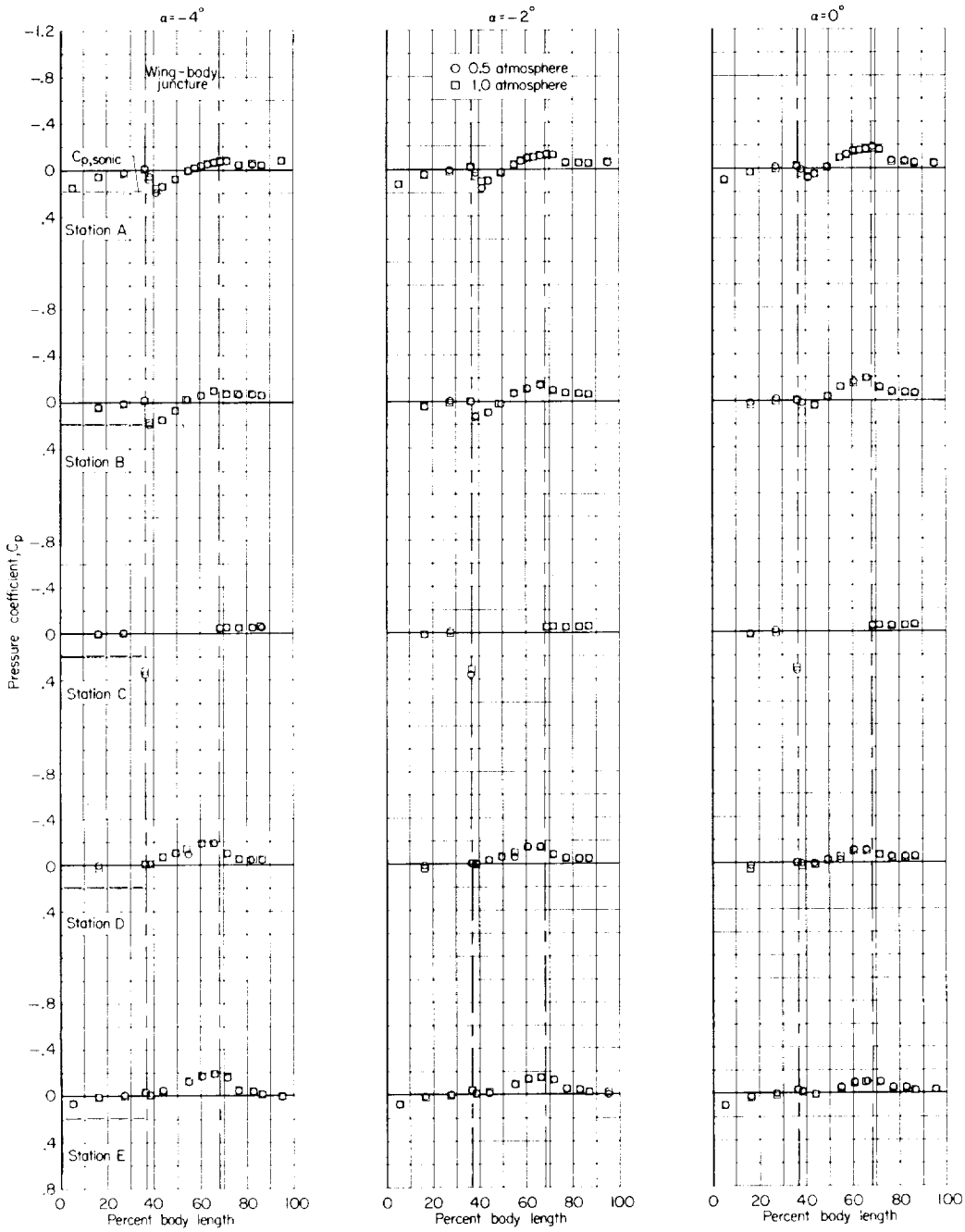
Figure 5.- Continued.



(f) M = 1.125.

Figure 5.- Continued.

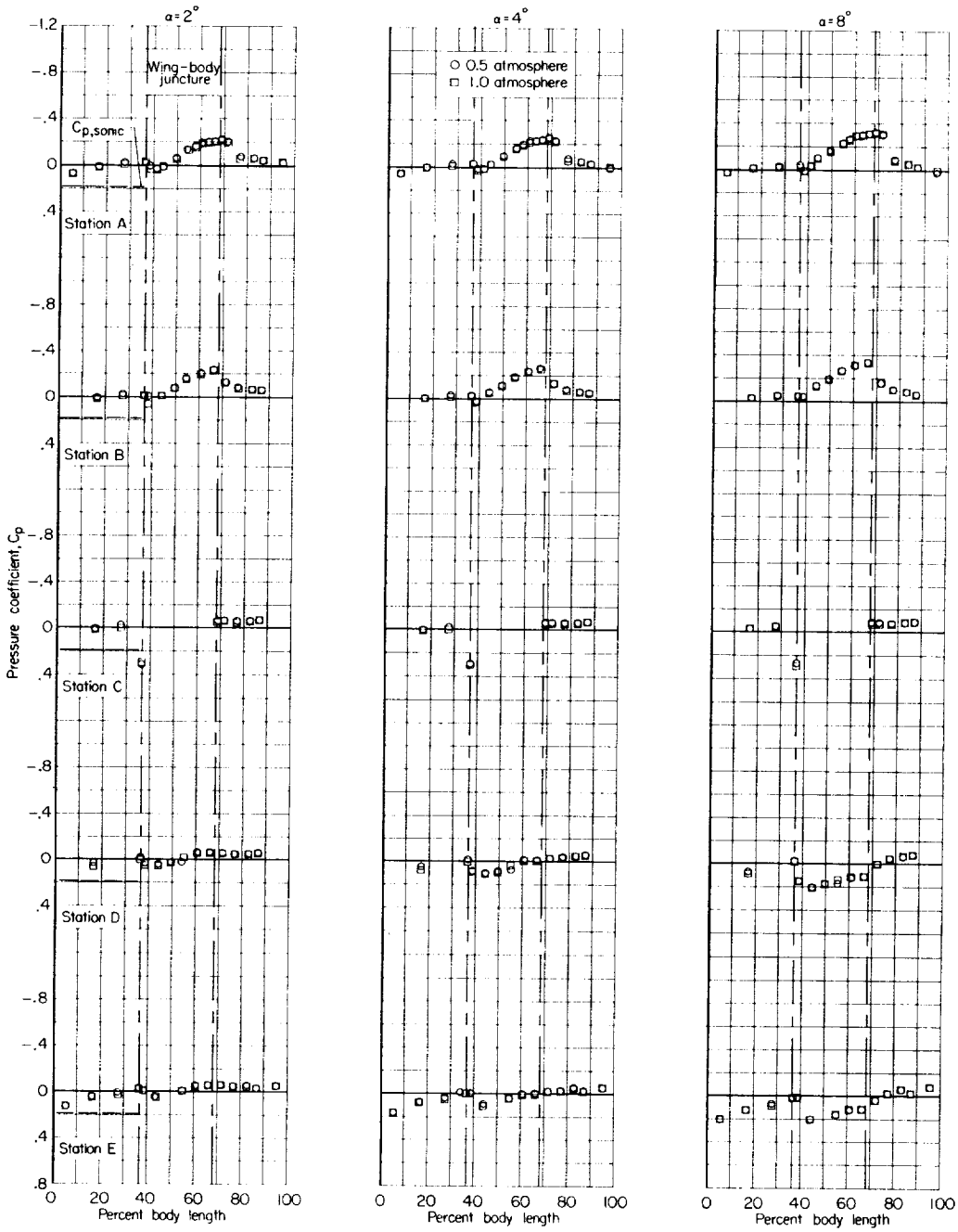


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Figure 5.- Continued.

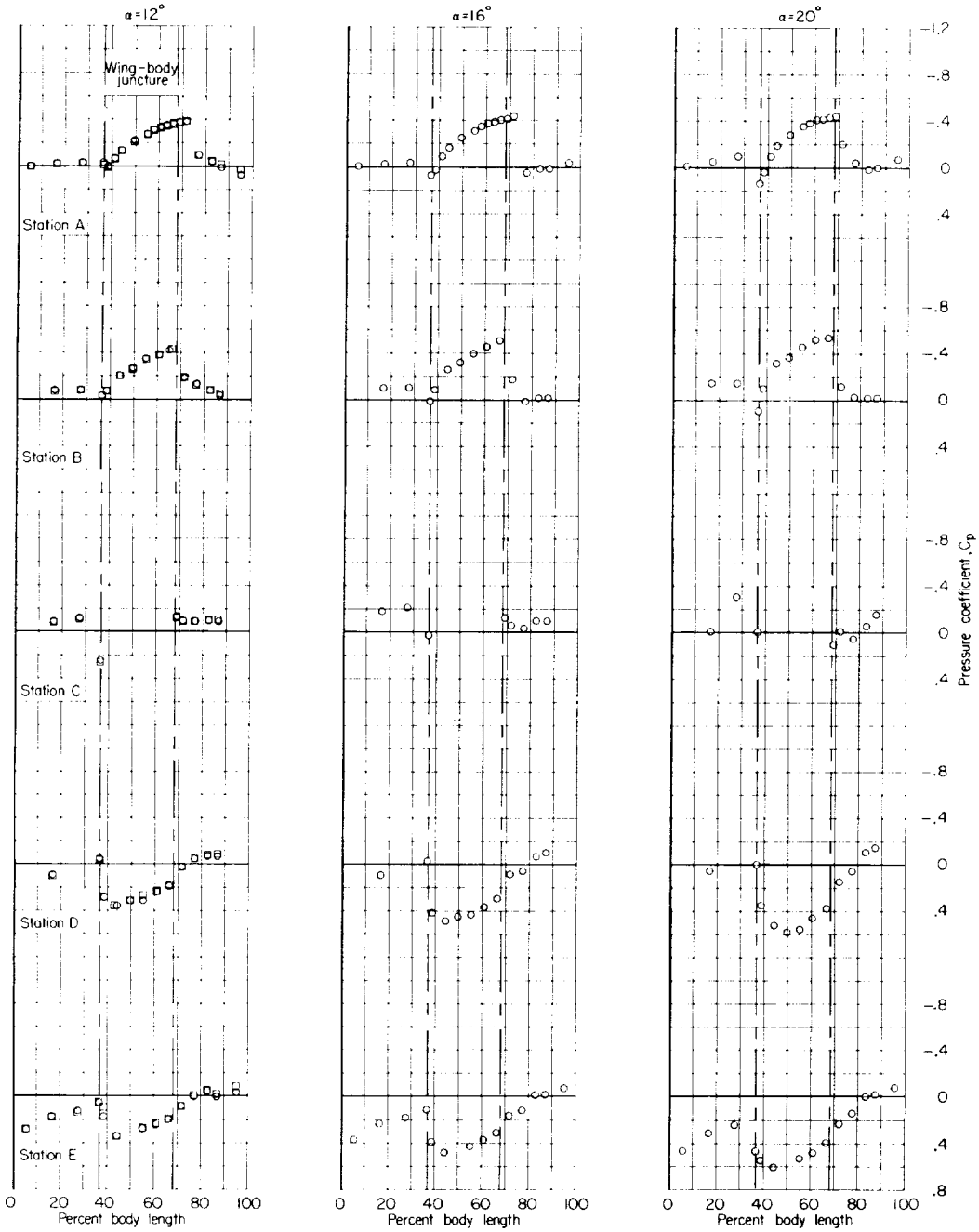

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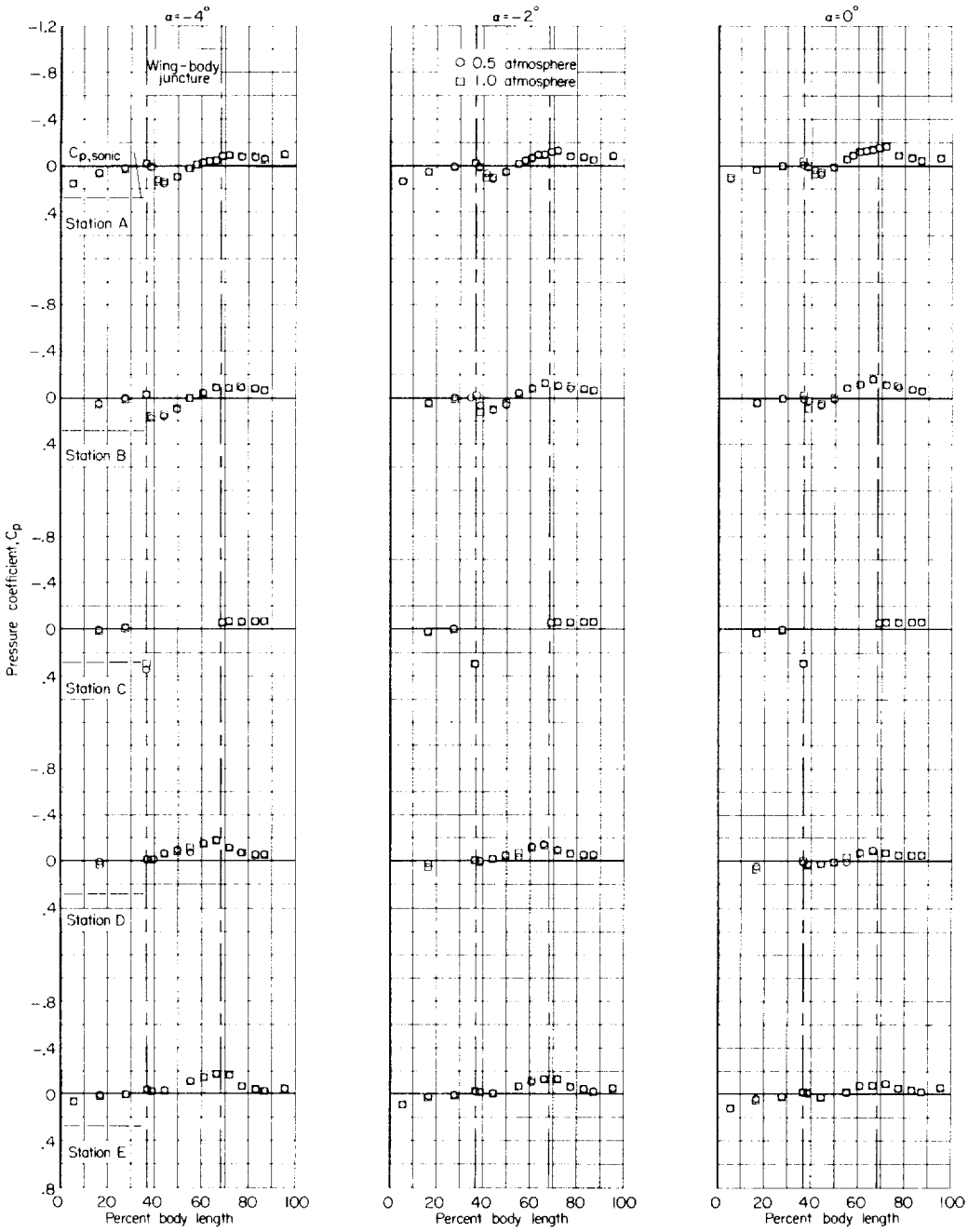
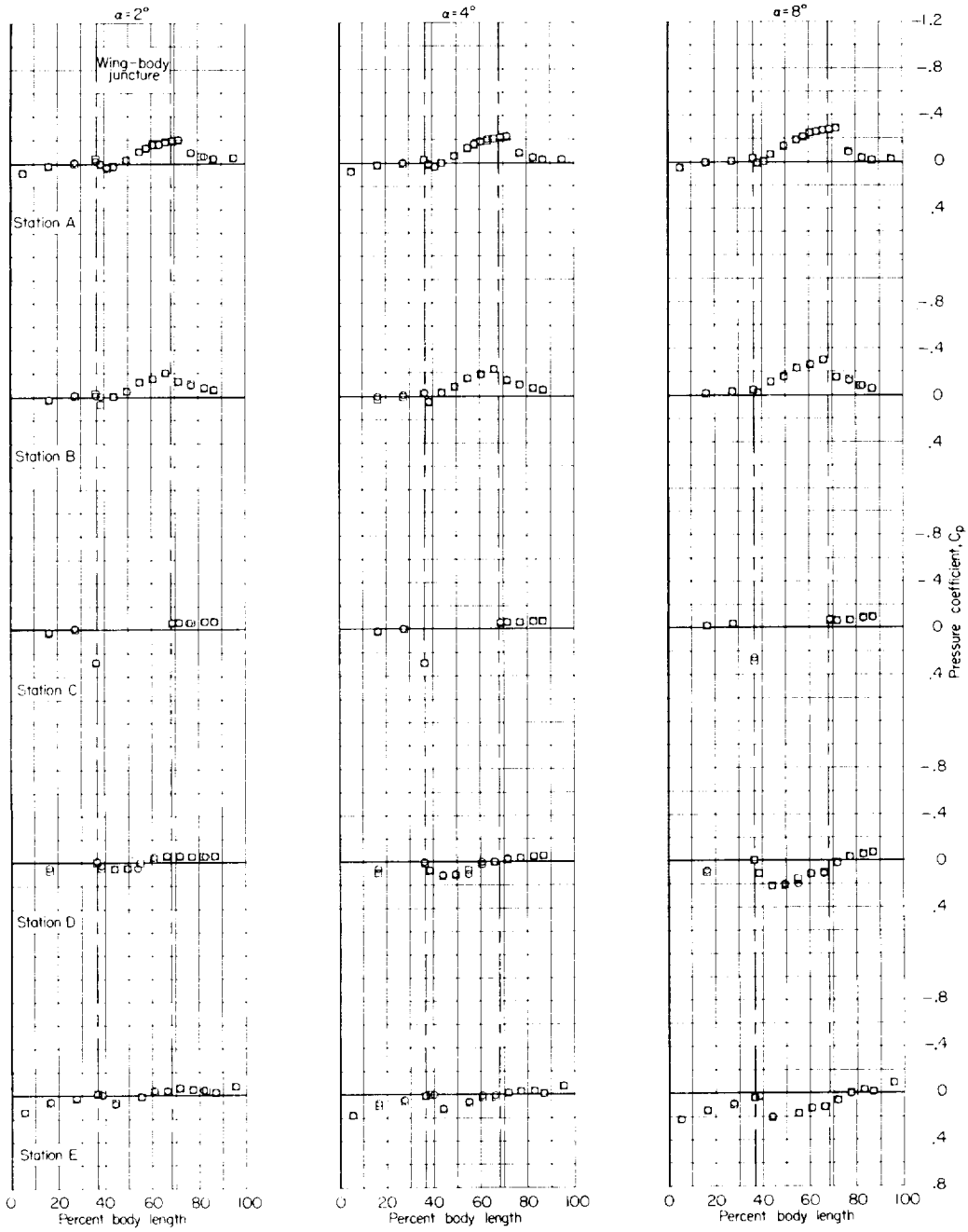
Figure 5.- Continued.



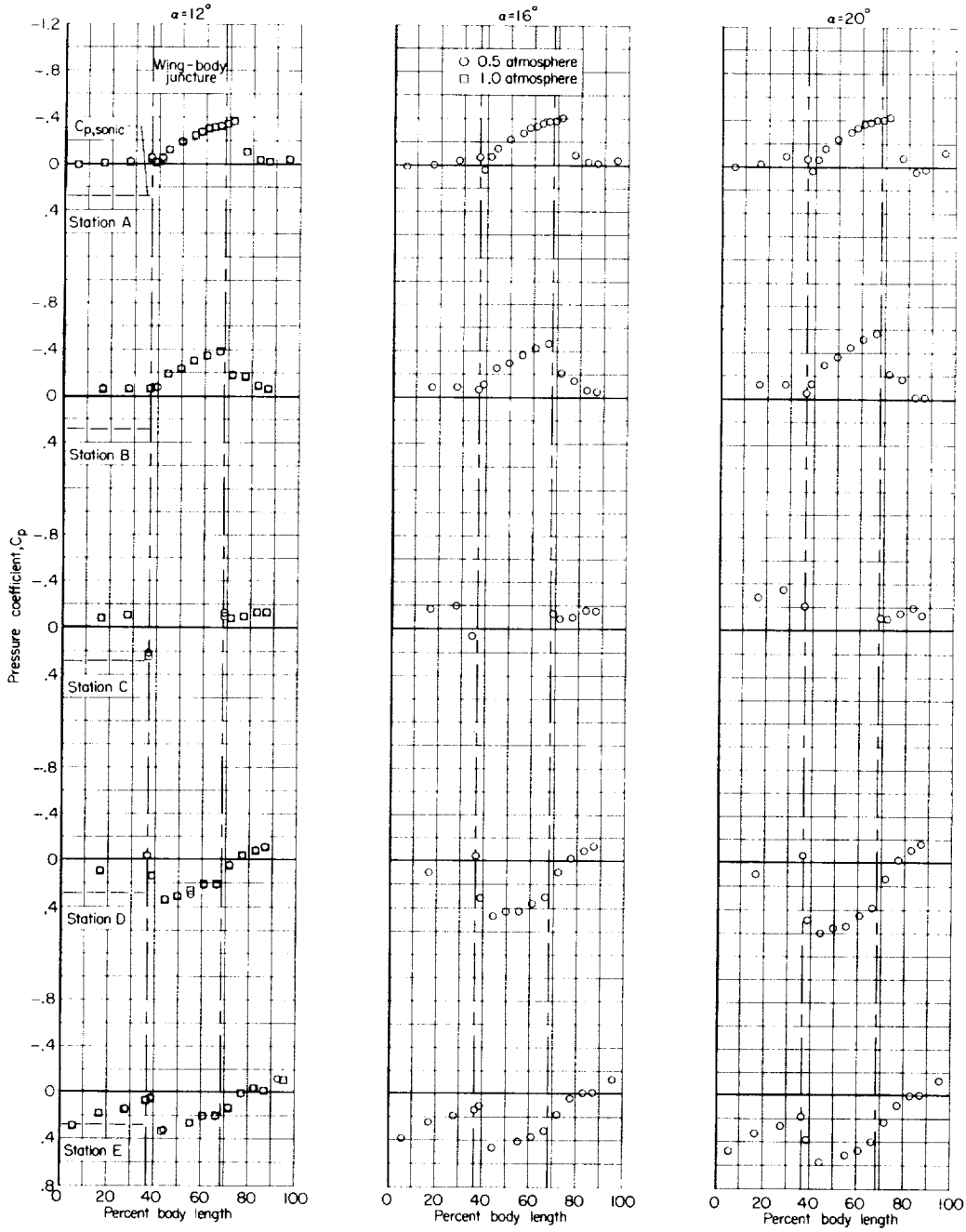
(g) M = 1.200.

Figure 5.- Continued.



(g) Continued.

Figure 5.- Continued.



(g) Concluded.

Figure 5.- Concluded.

NASA - Langley Field, Va. L-301

NASA MEMO 5-12-59L
National Aeronautics and Space Administration.
BASIC PRESSURE MEASUREMENTS AT TRANSONIC SPEEDS ON A THIN 45° SWEPTBACK HIGHLY TAPERED WING WITH SYSTEMATIC SPANWISE TWIST VARIATIONS. WING WITH CUBIC SPANWISE TWIST VARIATION. John P. Mugler, Jr. June 1959. 148p. diagrs., tabs. (NASA MEMORANDUM 5-12-59L)

Pressure data are presented which were obtained in the Langley 8-foot transonic pressure tunnel. Tests were made at Mach numbers from 0.800 to 1.200 at stagnation pressures of 1.0 and 0.5 atmosphere and at angles of attack from -4° to 12°. The wing has 45° sweepback of the quarter-chord line, taper ratio of 0.15, and an aspect ratio of 4. The wing was cambered and had a thickened root section.

Copies obtainable from NASA, Washington

1. Wings, Complete - Taper and Twist (1.2.2.2.4)
2. Mach Number Effects - Complete Wings (1.2.2.6)
3. Wing-Fuselage Combinations - Airplanes (1.7.1.1.1)
4. Loads, Steady - Wings (4.1.1.1.1)
5. Loads - Aeroelasticity (4.1.1.5)

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