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MEMORANDUM

for

UNITED STATES AIR FORCE

AN EXPERIMENTAL INVESTIGATION OF THE PRESSURE DISTRIBUTION ON A 1/15-SCALE MODEL OF THE LOCKHEED WS-117L VEHICLE PLUS BOOSTER "B" AT MACH NUMBERS FROM 0.70 TO 1.45 (COORD. NO. AF-AM-163)

By Russell E. Fahey and Ralph D. Marker

Ames Research Center
Moffett Field, Calif.

SERVICE REPORT

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NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

WASHINGTON

March 1959

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NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

NASA MEMO 3-12-59A

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ABSTRACT

Results obtained with two nose shapes tested at a Reynolds number per foot of 5×10^6 at angles of attack from -4° to $+10^\circ$ at 0° angle of sideslip are presented in tabulated pressure coefficient form without analysis.

INDEX HEADINGS

Missiles, Specific Types 1.7.2.2

Loads, Aerodynamic 4.1.1

*Title, [redacted]

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AN EXPERIMENTAL INVESTIGATION OF THE PRESSURE DISTRIBUTION ON A 1/15-SCALE MODEL OF THE LOCKHEED WS-117L VEHICLE PLUS BOOSTER "B" AT MACH NUMBERS FROM 0.70 TO 1.45 (COORD. NO. AF-AM-163)*

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SUMMARY

An investigation of the pressure distribution on a 1/15-scale model of the Lockheed WS-117L vehicle plus Booster "B" with two nose shapes has been conducted.

Measurements of static pressure were made on the surface of the vehicle and forward portion of the booster body at a Reynolds number per foot of 5×10^6 through a Mach number range of 0.70 to 1.45 and an angle-of-attack range of -4° to $+10^\circ$ at 0° angle of sideslip.

The results are presented in tabulated pressure-coefficient form without analysis in order to expedite publication.

INTRODUCTION

At the request of the United States Air Force, an investigation has been conducted in the Ames Unitary Plan wind tunnel to determine the static longitudinal stability and loading characteristics of the Lockheed WS-117L vehicle plus Booster "B". The experimental data will provide information for control systems analysis, trajectory calculations, and vehicle structural loadings for the Mach number range of 0.70 to 3.50.

This report presents the results of the pressure distribution tests conducted in the 11- by 11-foot transonic test section on a 1/15-scale model of the Lockheed WS-117L vehicle plus Booster "B" in the Mach number range of 0.70 to 1.45. Reference 1 presents the results of the pressure distribution tests in the Mach number range of 1.55 to 2.35.

*Title, [redacted]

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The results of the static longitudinal stability tests in the Mach number ranges of 0.70 to 1.45, 1.55 to 2.35, and 2.5 to 3.5 are presented in references 2, 3, and 4, respectively.

NOTATION

C_p pressure coefficient, $\frac{p_l - p_\infty}{q_\infty}$

M_∞ free-stream Mach number

R Reynolds number

l reference model length, 75.696 in.

p_∞ free-stream static pressure, lb/sq ft

p_l local pressure at orifice, lb/sq ft

q_∞ free-stream dynamic pressure, lb/sq ft

x distance from fuselage station 0 measured along model longitudinal center line, in.

α angle of attack, deg

β angle of sideslip, deg

θ angular location of orifices about model vertical center line, deg ($\theta = 0^\circ$ at bottom of model increasing clockwise facing upstream)

Configuration Symbols

IB WS-117L vehicle plus Booster "B"

N nose configuration, further identified by subscripts as shown in figure 3

APPARATUS AND TESTS

Tunnel

The tests were conducted in the 11- by 11-foot transonic wind tunnel, which is a closed circuit, variable-pressure type with perforated

[redacted]

test section walls as described in reference 5. Mach number and stagnation pressure are continuously variable from 0.50 to 1.45 and 2 to 35 pounds per square inch absolute, respectively. The nozzle is a convergent-divergent type with flexible side walls to provide the variable geometry required to produce the range of Mach numbers.

The test section is 11 feet square in cross section and 22 feet in length. A pressure-tight plenum chamber serves to equalize the pressure on the four enclosed perforated walls. A slight divergence of the test-section side walls approximately compensates for boundary-layer growth.

Model

The model is shown installed in the test section in figure 1. The general arrangement of the model, the nose shape details, and the pressure orifice locations are shown in figures 2, 3, and 4, respectively.

Tests

Model static pressures were measured by means of Ames precision strain-gage differential pressure transducers as described in reference 6. Photographically recorded manometer boards were utilized as a check on the automatic pressure reading equipment.

The tests were conducted at Mach numbers of 0.70, 0.90, 1.00, 1.10, 1.20, 1.30, and 1.45 through an angle-of-attack range of -4° to $+10^{\circ}$ at 0° angle of sideslip. The Reynolds number was 5×10^{6} per foot. Transition was not fixed on the model.

Corrections to Data

Tunnel air-stream surveys show that the flow in the porous region of the test section is generally uniform and that no significant pressure gradients exist. Therefore, no longitudinal buoyancy or air-stream angle corrections were applied to the data.

For the size of model tested wall interference is believed to be negligible necessitating no correction to the data.

A blank space in the presented tabulated data represents a plugged orifice condition.

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RESULTS

The results are presented in tabulated pressure-coefficient form as listed below without analysis in order to expedite publication.

Table no.	Configuration	Mach no.
1	IBN_5	0.70
2	↓	.90
3	↓	1.00
4	↓	1.10
5	↓	1.20
6	↓	1.30
7	↓	1.45
8	IBN_7	.70
9	↓	.90
10	↓	1.00
11	↓	1.10
12	↓	1.20
13	↓	1.28
14	↓	1.43

Ames Research Center
National Aeronautics and Space Administration
Moffett Field, Calif., Jan. 14, 1959

REFERENCES

1. Martin, Norman J.: An Investigation of the Pressure Distribution on a 1/15-Scale Model of the Lockheed WS-117L Vehicle Plus Booster "B" at Mach Numbers From 1.55 to 2.35 (COORD. NO. AF-AM-163). NASA MEMO 3-13-59A, 1959.

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3. Martin, Norman J.: An Experimental Investigation of the Aerodynamic Characteristics of a 1/15-Scale Model of the Lockheed WS-117L Vehicle Plus Booster "B" at Mach Numbers From 1.55 to 2.35 (COORD. NO. AF-AM-163). NASA MEMO 10-11-58A, 1958.

4. Bright, Loren G.: An Investigation of the Aerodynamic Characteristics of a 1/15-Scale Model of the Lockheed WS-117L Vehicle With Booster "B" at Mach Numbers From 2.49 to 3.53 (COORD. NO. AF-AM-163). NASA MEMO 10-12-58A, 1958.

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5. Hunton, Lynn W.: Effects of Fixing Transition on the Transonic Aerodynamic Characteristics of a Wing-Body Configuration at Reynolds Numbers From 2.4 to 12 Million. NACA TN 4279, 1958.

6. Dimeff, John: A Survey of New Developments in Pressure Measuring Techniques in the NACA. AGARD Rep. 166, 1958.

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TABLE I.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE No. FOR M_∞=0.70

(a) α = -4.2°

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.223	0.540	0.551	0.584	0.637	0.683	0.825	0.826	0.793	0.729	0.689
0.226	-0.193	-0.182	-0.138	-0.070	-0.022	0.173	0.166	0.112	0.061	-0.012
0.238	0.164	0.169	0.170	0.198	0.221	0.318	0.271	0.293	0.249	0.229
0.251	0.119	0.094	0.114	0.129	0.157	0.246	0.262	0.211	0.165	0.130
0.269	-0.028	-0.047	-0.034	-0.013	0.007	0.100	0.120	0.081	0.042	-0.002
0.286	-0.359	-0.331	-0.347	-0.334	-0.313	-0.240	-0.242	-0.274	-0.294	-0.328
0.292	-0.702	-0.698	-0.698	-0.690		-0.580	-0.582	-0.613	-0.647	-0.671
0.322	-0.038	-0.039	-0.016	-0.048	-0.048	-0.027	-0.029	-0.033	-0.013	0.040
0.352	0.029	0.101	0.042	0.008	0.005	0.028	0.021	0.009	0.007	0.003
0.381	0.127	0.126		0.117	0.121	0.161	0.158	0.149	0.135	0.105
0.392	0.070	0.053	0.059	0.062	0.066	0.110	0.108	0.092	0.075	0.048
0.402	-0.020	-0.030	-0.024	-0.027	-0.022	0.013	-0.009	-0.004	-0.014	-0.014
0.447	0.015	0.015	0.009	0.012	0.013		0.055	0.028	0.019	-0.000
0.492	0.013	0.013	0.008	0.009	0.003	0.045	0.042	0.034	0.021	0.008
0.529	0.004	0.007	0.003	-0.000	0.002	0.035	0.036	0.012	0.008	-0.007
0.592	-0.003	-0.003	-0.007	-0.006	-0.009	0.029	0.025	0.020	0.003	-0.008
0.655	-0.112	-0.111	-0.111	-0.115	-0.124	-0.104	-0.071	-0.114	-0.114	-0.126
0.661	-0.124	-0.182	-0.123	-0.131	-0.128	-0.143		-0.135	-0.137	-0.124
0.687	-0.280	0.032	-0.025	-0.049	-0.034	-0.269	0.059	-0.021		-0.042

(b) α = -2.2°

0.223	0.621	0.629	0.642	0.667	0.694	0.760	0.767	0.751	0.717	0.699
0.226	-0.095	-0.093	-0.071	-0.036	-0.016	0.082	0.081	0.054	0.023	-0.004
0.238	0.202	0.204	0.201	0.220	0.231	0.276	0.239	0.269	0.246	0.241
0.251	0.153	0.128	0.145	0.150	0.166	0.208	0.227	0.190	0.164	0.143
0.269	0.003	-0.014	-0.005	0.006	0.018	0.064	0.080	0.059	0.042	0.008
0.286	-0.337	-0.309	-0.328	-0.322	-0.308	-0.273	-0.272	-0.293	-0.297	-0.322
0.292	-0.694	-0.688	-0.683	-0.675		-0.624	-0.621	-0.636	-0.650	-0.664
0.322	-0.037	-0.024	-0.028	-0.042	-0.040	-0.036	-0.035	-0.036	-0.026	0.062
0.352	0.025	0.085	0.046	0.016	0.014	0.023	0.017	0.012	0.017	0.047
0.381	0.134	0.133		0.129	0.131	0.151	0.150	0.148	0.141	0.118
0.392	0.074	0.058	0.067	0.073	0.075	0.094	0.095	0.087	0.081	0.055
0.402	-0.019	-0.028	-0.026	-0.023	-0.013	0.003	-0.004	-0.004	-0.006	-0.007
0.447	0.020	0.019	0.016	0.022	0.023		0.033	0.026	0.026	0.014
0.492	0.018	0.018	0.016	0.020	0.016	0.034	0.034	0.032	0.027	0.018
0.529	0.004	0.013	0.010	0.011	0.013	0.026	0.027	0.021	0.014	-0.001
0.592	0.001	0.002	-0.000	0.002	0.003	0.016	0.017	0.015	0.004	-0.004
0.655	-0.116	-0.116	-0.110	-0.106	-0.115	-0.113	-0.074	-0.112	-0.110	-0.114
0.661	-0.132	-0.198	-0.123	-0.122	-0.117	-0.136		-0.132	-0.126	-0.110
0.687	-0.014	0.037	-0.023	-0.038	-0.027	-0.010	0.051	-0.021		-0.297

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TABLE I.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_2 FOR M_∞=0.70 - Continued

(c) α = -0.1°

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.223	0.691	0.694	0.693	0.690	0.691	0.689	0.702	0.701	0.696	0.697
0.226	-0.011	-0.011	-0.007	-0.013	-0.021	-0.013	-0.007	-0.013	-0.009	-0.008
0.238	0.236	0.237	0.226	0.230	0.230	0.233	0.201	0.241	0.234	0.242
0.251	0.185	0.158	0.167	0.161	0.162	0.164	0.188	0.163	0.151	0.142
0.269	0.033	-0.000	0.005	0.016	0.017	0.026	0.039	0.030	0.026	0.009
0.286	-0.317	-0.289	-0.315	-0.318	-0.311	-0.305	-0.302	-0.314	-0.305	-0.323
0.292	-0.678	-0.675	-0.669	-0.669		-0.660	-0.655	-0.763	-0.660	-0.664
0.322	-0.034	-0.007	-0.044	-0.043	-0.039	-0.043	-0.041	-0.042	-0.047	0.023
0.352	0.022	0.062	0.045	0.017	0.015	0.018	0.013	0.011	0.023	0.086
0.381	0.138	0.132		0.133	0.133	0.140	0.140	0.142	0.142	0.126
0.392	0.081	0.062	0.075	0.077	0.076	0.081	0.083	0.080	0.079	0.059
0.402	-0.018	-0.028	-0.026	-0.017	-0.013	-0.007	-0.010	-0.011	-0.009	-0.005
0.447	0.023	0.020	0.019	0.024	0.022		0.020	0.017	0.021	0.013
0.492	0.019	0.019	0.017	0.020	0.015	0.019	0.020	0.022	0.023	0.019
0.529	0.011	0.014	0.013	0.012	0.011	0.011	0.013	0.011	0.010	0.008
0.592	0.004	0.004	-0.002	0.006	0.005	0.005	0.019	-0.002	0.013	0.002
0.655	-0.120	-0.116	-0.113	-0.105	-0.113	-0.119	-0.082	-0.121	-0.112	-0.112
0.661	-0.138	-0.205	-0.129	-0.121	-0.123	-0.135		-0.130	-0.124	-0.115
0.687	-0.280	0.036	-0.021	-0.038	-0.025	-0.281	0.038	-0.026		-0.032

(d) α = 3.9°

0.223	0.825	0.819	0.784	0.727	0.684	0.543	0.558	0.596	0.642	0.685
0.226	0.170	0.156	0.114	0.033	-0.028	-0.190	-0.178	-0.145	-0.071	-0.013
0.238	0.317	0.312	0.279	0.244	0.218	0.162	0.139	0.183	0.201	0.229
0.251	0.265	0.230	0.217	0.173	0.150	0.097	0.126	0.110	0.120	0.131
0.269	0.111	0.083	0.065	0.030	0.009	-0.034	-0.024	-0.021	-0.009	-0.004
0.286	-0.254	-0.232	-0.278	-0.308	-0.320	-0.350	-0.347	-0.352	-0.324	-0.329
0.292	-0.611	-0.613	-0.633	-0.667		-0.696	-0.692	-0.695	-0.691	-0.681
0.322	-0.019	0.038	-0.077	-0.056	-0.049	-0.046	-0.044	-0.054	-0.098	0.013
0.352	0.028	0.062	0.067	0.014	0.005	0.023	0.013	0.009	0.036	0.030
0.381	0.158	0.146		0.128	0.122	0.129	0.127	0.128	0.135	0.105
0.392	0.107	0.081	0.089	0.073	0.065	0.069	0.069	0.064	0.071	0.049
0.402	-0.007	-0.018	-0.021	-0.027	-0.027	-0.013	-0.018	-0.020	-0.019	-0.014
0.447	0.046	0.038	0.030	0.020	0.012		0.011	0.006	0.025	0.003
0.492	0.042	0.037	0.026	0.016	0.005	0.008	0.011	0.011	0.008	0.009
0.529	0.034	0.033	0.021	0.004	0.002	0.003	0.002	-0.000	-0.002	-0.003
0.592	0.027	0.022	0.018	0.002	-0.006	-0.007	-0.008	-0.003	-0.008	-0.010
0.655	-0.112	-0.113	-0.120	-0.117	-0.128	-0.116	-0.093	-0.114	-0.117	-0.125
0.661	-0.135	-0.207	-0.137	-0.137	-0.131	-0.123		-0.130	-0.135	-0.126
0.687	-0.005	0.054	-0.020	-0.048	-0.036	-0.018	0.021	-0.033		-0.044

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TABLE I.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE No. FOR M_∞ = 0.70 - Continued

(e) α = 6.0°

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.223	0.883	0.873	0.821	0.734	0.671	0.461	0.483	0.535	0.607	0.670
0.226	0.260	0.238	0.169	0.051	-0.039	-0.283	-0.271	-0.218	-0.110	-0.025
0.238	0.359	0.347	0.295	0.240	0.199	0.128	0.107	0.149	0.171	0.210
0.251	0.305	0.266	0.234	0.170	0.130	0.065	0.094	0.077	0.095	0.112
0.269	0.151	0.119	0.086	0.029	-0.008	-0.063	-0.053	-0.051	-0.049	-0.029
0.286	-0.204	-0.187	-0.253	-0.311	-0.338	-0.378	-0.375	-0.382	-0.350	-0.347
0.292	-0.552	-0.562	-0.596	-0.647		-0.679	-0.680	-0.704	-0.707	-0.684
0.322	0.006	0.055	-0.111	-0.086	-0.058	-0.037	-0.038	-0.067	-0.120	-0.043
0.352	0.026	0.072	0.075	-0.011	-0.026	0.037	0.023	0.004	0.039	-0.009
0.381	0.160	0.147		0.100	0.107	0.124	0.121	0.118	0.119	0.097
0.392	0.124	0.092	0.090	0.061	0.048	0.064	0.062	0.052	0.057	0.031
0.402	0.008	-0.019	-0.019	-0.035	-0.042	-0.012	-0.281	-0.032	-0.035	-0.027
0.447	0.064	0.052	0.026	0.004	-0.007		0.006	-0.005	-0.011	-0.016
0.492	0.064	0.054	0.027	0.001	-0.009	-0.007	0.007	0.001	-0.011	-0.012
0.529	0.059	0.052	0.024	-0.008	-0.019	0.002	-0.000	-0.012	-0.024	-0.019
0.592	0.053	0.043	0.015	-0.013	-0.028	-0.010	-0.011	-0.018	-0.032	-0.028
0.655	-0.081	-0.108	-0.125	-0.138	-0.157	-0.108	-0.086	-0.120	-0.145	-0.155
0.661	-0.132	-0.198	-0.142	-0.163	-0.159	-0.116		-0.128	-0.162	-0.157
0.687	0.016	0.070	-0.023	-0.060	-0.065	-0.011	0.022	-0.034		-0.077

(f) α = 8.0°

0.223	0.939	0.923	0.851	0.737	0.652	0.375	0.403	0.465	0.563	0.649
0.226	0.346	0.317	0.226	0.063	-0.056	-0.380	-0.367	-0.297	-0.157	-0.042
0.238	0.401	0.376	0.314	0.226	0.174	0.093	0.072	0.108	0.136	0.184
0.251	0.346	0.300	0.249	0.160	0.102	0.032	0.061	0.039	0.060	0.086
0.269	0.192	0.154	0.102	0.010	-0.049	-0.094	-0.085	-0.086	-0.110	-0.074
0.286	-0.152	-0.143	-0.236	-0.331	-0.374	-0.407	-0.408	-0.421	-0.392	-0.382
0.292	-0.412	-0.439	-0.513	-0.638		-0.660	-0.667	-0.816	-0.727	-0.696
0.322	0.024	0.085	-0.140	-0.119	-0.087	-0.035	-0.040	-0.084	-0.212	-0.125
0.352	0.057	0.070	0.102	-0.041	-0.061	0.039	0.021	-0.002	0.025	-0.030
0.381	0.200	0.175		0.069	0.083	0.119	0.114	0.107	0.096	0.074
0.392	0.142	0.102	0.087	0.042	0.023	0.060	0.055	0.037	0.025	-0.000
0.402	0.028	0.005	-0.020	-0.063	-0.070	-0.013	-0.026	-0.049	-0.069	-0.068
0.447	0.083	0.066	0.020	-0.022	-0.041		-0.003	-0.025	-0.045	-0.041
0.492	0.087	0.069	0.019	-0.034	-0.034	0.002	-0.004	-0.021	-0.052	-0.031
0.529	0.084	0.068	0.016	-0.046	-0.038	-0.007	-0.014	-0.029	-0.067	-0.046
0.592	0.079	0.060	0.006	-0.048	-0.070	-0.013	-0.027	-0.029	-0.048	-0.077
0.655	-0.057	-0.109	-0.132	-0.169	-0.193	-0.112	-0.115	-0.128	-0.176	-0.190
0.661	-0.093	-0.198	-0.149	-0.193	-0.193	-0.128		-0.134	-0.191	-0.168
0.687	0.025	0.070	-0.036	-0.080	-0.097	-0.019	0.011	-0.042		-0.372

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[redacted]

TABLE I.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE No. FOR $M_\infty = 0.70$ - Concluded

(g) $\alpha = 10.0°$

Axial position x/l	Radial position, θ, deg									
	0	20	40	72	90	180	200	225	252	270
0.223	0.987	0.965	0.876	0.733	0.625	0.281	0.314	0.389	0.513	0.620
0.226	0.429	0.392	0.272	0.070	-0.079	-0.482	-0.466	-0.381	-0.211	-0.069
0.238	0.440	0.408	0.327	0.204	0.135	0.056	0.038	0.062	0.088	0.140
0.251	0.386	0.334	0.259	0.139	0.063	-0.001	0.027	-0.008	0.017	0.047
0.269	0.231	0.187	0.115	0.001	-0.060	-0.131	-0.117	-0.126	-0.122	-0.086
0.286	-0.157	-0.151	-0.245	-0.342	-0.386	-0.418	-0.419	-0.432	-0.403	-0.393
0.292	-0.423	-0.445	-0.528	-0.657		-0.674	-0.681	-0.832	-0.744	-0.714
0.322	0.015	0.111	-0.150	-0.129	-0.126	-0.043	-0.049	-0.093	-0.222	-0.232
0.352	0.049	0.061	0.094	-0.051	-0.071	0.030	0.012	-0.010	0.016	-0.041
0.381	0.193	0.168		.061	0.046	0.110	0.102	0.090	0.057	0.038
0.392	0.156	0.109	0.072	0.011	-0.013	0.052	0.042	0.017	-0.022	-0.039
0.402	0.039	0.009	-0.033	-0.092	-0.107	-0.021	-0.030	-0.068	-0.090	-0.096
0.447	0.088	0.066	0.010	-0.044	-0.063		-0.013	-0.040	-0.063	-0.080
0.492	0.083	0.062	0.009	-0.048	-0.070	-0.007	-0.013	-0.032	-0.064	-0.069
0.529	0.077	0.060	0.007	-0.056	-0.074	-0.015	-0.023	-0.038	-0.077	-0.081
0.592	0.070	0.051	-0.002	-0.062	-0.081	-0.023	-0.037	-0.042	-0.080	-0.086
0.655	-0.066	-0.118	-0.142	-0.179	-0.203	-0.121	-0.124	-0.139	-0.186	-0.201
0.661	-0.106	-0.208	-0.158	-0.203	-0.203	-0.138		-0.001	-0.201	-0.201
0.687	0.016	0.062	-0.044	-0.112	-0.371	-0.025	0.002	-0.052		-0.120

[redacted]

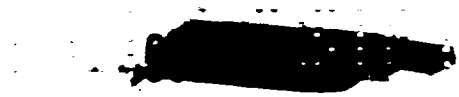

TABLE II.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE No. FOR M_∞ = [illegible]

(a) α = [illegible]°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	0.653	0.665	0.695	0.740	0.786	0.917	0.917	0.887	0.828	0.790
0.226	-0.180	-0.174	-0.120	-0.045	0.006	0.213	0.206	0.153	0.080	0.004
0.238	0.227	0.233	0.235	0.262	0.285	0.380	0.338	0.356	0.311	0.292
0.251	0.185	0.163	0.183	0.197	0.227	0.314	0.338	0.280	0.234	0.200
0.269	0.050	0.035	0.049	0.069	0.091	0.184	0.204	0.165	0.129	0.081
0.286	-0.126	-0.105	-0.115	-0.103	-0.085	-0.018	-0.020	-0.047	-0.068	-0.096
0.292	-1.009	-1.011	-1.011	-0.985		-0.964	-0.902	-1.031	-0.992	-0.999
0.322	-0.165	-0.154	-0.086	-0.048	-0.022	0.030	0.026	0.020	0.037	0.253
0.352	0.076	0.139	0.068	0.040	0.035	0.047	0.041	0.029	0.019	0.074
0.381	0.153	0.161		0.162	0.164	0.202	0.201	0.193	0.177	0.148
0.392	0.092	0.074	0.084	0.088	0.093	0.139	0.137	0.121	0.101	0.068
0.402	-0.017	-0.034	-0.037	-0.042	-0.035	0.002	-0.004	-0.015	-0.024	-0.022
0.447	0.014	0.012	0.009	0.012	0.012		0.040	0.026	0.018	-0.011
0.492	0.010	0.009	0.007	0.008	0.004	0.044	0.042	0.033	0.019	0.007
0.529	-0.003	0.011	-0.000	-0.003	-0.001	0.032	0.031	0.018	-0.011	-0.006
0.592	-0.012	-0.014	-0.016	-0.014	-0.012	0.024	0.022	0.013	-0.003	-0.020
0.655	-0.179	-0.351	-0.179	-0.184	-0.192	-0.124	-0.008	-0.155	-0.182	-0.199
0.661	-0.213	-0.275	-0.207	-0.211	-0.212	-0.367		-0.373	-0.268	-0.216
0.687	-0.029	0.011	-0.041	-0.061	-0.055	-0.011	0.057	-0.028		-0.059

(b) α = [illegible]°

0.223	0.725	0.732	0.749	0.771	0.795	0.858	0.861	0.849	0.815	0.798
0.226	-0.072	-0.071	-0.040	-0.004	0.020	0.122	0.119	0.091	0.058	0.028
0.238	0.264	0.269	0.267	0.284	0.296	0.342	0.305	0.336	0.310	0.305
0.251	0.223	0.199	0.216	0.221	0.238	0.279	0.306	0.262	0.236	0.214
0.269	0.089	0.070	0.081	0.094	0.106	0.152	0.168	0.146	0.128	0.095
0.286	-0.097	-0.076	-0.089	-0.083	-0.071	-0.040	-0.039	-0.056	-0.062	-0.082
0.292	-0.997	-0.997	-0.995	-0.972		-0.950	-0.891	-1.019	-0.984	-0.986
0.322	-0.080	-0.055	-0.073	-0.032	-0.012	0.020	0.023	0.018	0.020	0.276
0.352	0.068	0.129	0.076	0.056	0.053	0.054	0.050	0.045	0.041	0.051
0.381	0.173	0.179		0.147	0.178	0.196	0.197	0.196	0.184	0.163
0.392	0.105	0.086	0.097	0.105	0.107	0.128	0.129	0.121	0.111	0.082
0.402	-0.020	-0.033	-0.026	-0.024	-0.019	-0.002	-0.006	-0.010	-0.010	-0.008
0.447	0.021	0.021	0.019	0.026	0.026		0.034	0.027	0.028	0.016
0.492	0.020	0.020	0.019	0.023	0.019	0.037	0.037	0.035	0.030	0.022
0.529	0.010	0.012	-0.001	0.013	0.012	0.024	0.025	0.019	0.012	0.002
0.592	-0.002	-0.002	-0.002	0.001	0.001	0.015	0.015	0.013	0.006	-0.001
0.655	-0.174	-0.090	-0.169	-0.166	-0.180	-0.139	-0.021	-0.169	-0.167	-0.174
0.661	-0.243	-0.630	-0.212	-0.195	-0.186	-0.341		-0.301	-0.207	-0.181
0.687	-0.025	0.030	-0.030	-0.046	-0.039	-0.012	0.052	-0.024		-0.045

[redacted]

TABLE II.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_5 FOR M_∞=0.90 - Continued

(c) α = -0.1°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	0.789	0.792	0.791	0.791	0.792	0.791	0.798	0.801	0.796	0.796
0.226	0.025	0.018	0.031	0.021	0.006	0.022	0.023	-0.000	0.023	0.025
0.238	0.298	0.299	0.293	0.296	0.296	0.299	0.269	0.306	0.299	0.305
0.251	0.254	0.229	0.238	0.231	0.234	0.236	0.270	0.235	0.223	0.214
0.269	0.120	0.100	0.104	0.105	0.107	0.115	0.127	0.119	0.115	0.095
0.286	-0.076	-0.056	-0.075	-0.077	-0.072	-0.067	-0.065	-0.074	-0.069	-0.082
0.292	-0.990	-0.992	-0.992	-0.973		-0.988	-0.893	-1.021	-0.992	-0.989
0.322	-0.020	0.008	-0.040	-0.025	-0.022	-0.028	-0.023	-0.021	-0.031	0.267
0.352	0.056	0.097	0.090	0.055	0.053	0.056	0.053	0.050	0.061	0.044
0.381	0.185	0.182		0.177	0.176	0.182	0.184	0.186	0.185	0.170
0.392	0.111	0.090	0.105	-0.228	0.107	0.113	0.115	0.111	0.113	0.087
0.402	-0.022	-0.033	-0.029	-0.236	-0.016	-0.010	-0.012	-0.230	-0.012	-0.007
0.447	0.026	0.024	0.024	0.029	0.027		0.024	0.021	0.026	0.017
0.492	0.023	0.023	0.021	0.024	0.018	0.024	0.024	0.026	0.026	0.022
0.529	0.002	0.015	0.013	-0.000	0.012	0.012	0.014	0.011	0.011	0.009
0.592	0.003	0.002	0.002	0.004	0.002	0.002	0.003	0.006	0.004	-0.000
0.655	-0.170	-0.078	-0.170	-0.167	-0.176	-0.167	-0.052	-0.168	-0.167	-0.174
0.661	-0.301	-0.727	-0.266	-0.202	-0.188	-0.291		-0.244	-0.201	-0.185
0.687	-0.022	0.035	-0.029	-0.048	-0.042	-0.025	0.028	-0.036		-0.050

(d) α = 1.9°

0.223	0.854	0.854	0.837	0.810	0.792	0.726	0.738	0.756	0.773	0.795
0.226	0.118	0.113	0.100	0.050	0.008	-0.071	-0.062	-0.047	-0.004	0.025
0.238	0.337	0.338	0.320	0.306	0.294	0.264	0.237	0.281	0.286	0.304
0.251	0.293	0.264	0.263	0.242	0.231	0.203	0.215	0.210	0.212	0.212
0.269	0.158	0.135	0.130	0.112	0.104	0.083	0.093	0.093	0.098	0.090
0.286	-0.053	-0.035	-0.061	-0.075	-0.077	-0.093	-0.090	-0.094	-0.080	-0.086
0.292	-0.981	-0.985	-0.988	-0.974		-1.002	-0.894	-1.023	-0.999	-0.991
0.322	0.016	0.053	-0.014	-0.011	-0.017	-0.066	-0.057	-0.041	-0.046	0.253
0.352	0.053	0.092	0.099	0.053	0.051	0.060	0.057	0.053	0.070	0.039
0.381	0.189	0.184		0.177	0.176	0.173	0.176	0.180	0.190	0.166
0.392	0.124	0.100	0.115	0.110	0.104	0.103	0.105	0.103	0.108	0.083
0.402	-0.017	-0.027	-0.031	-0.022	-0.021	-0.013	-0.012	-0.019	-0.020	-0.009
0.447	0.034	0.030	0.028	0.028	0.022		0.016	0.013	0.021	-0.001
0.492	0.030	0.028	0.025	0.023	0.014	0.016	0.017	0.019	0.022	0.017
0.529	0.019	0.020	0.017	0.005	0.009	0.005	0.006	0.005	0.006	0.005
0.592	0.024	0.010	0.003	-0.005	-0.006	-0.006	-0.005	-0.001	-0.000	-0.003
0.655	-0.152	-0.066	-0.168	-0.173	-0.181	-0.175	-0.075	-0.170	-0.166	-0.175
0.661	-0.339	-0.734	-0.295	-0.213	-0.191	-0.246		-0.226	-0.195	-0.181
0.687	-0.017	0.041	-0.026	-0.048	-0.042	-0.025	0.024	-0.036		-0.050

[redacted]

[redacted]

TABLE II.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_5 FOR M_∞ = 0.90 - Continued

(e) α = 3.9°

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.223	0.914	0.908	0.875	0.822	0.783	0.653	0.669	0.700	0.743	0.782
0.226	0.211	0.202	0.163	0.071	0.008	-0.181	-0.171	-0.129	-0.043	0.017
0.238	0.376	0.373	0.340	0.304	0.281	0.226	0.201	0.247	0.263	0.290
0.251	0.331	0.298	0.283	0.241	0.217	0.166	0.201	0.177	0.191	0.200
0.269	0.193	0.167	0.149	0.114	0.093	0.046	0.055	0.060	0.073	0.078
0.286	-0.030	-0.015	-0.052	-0.079	-0.091	-0.120	-0.117	-0.119	-0.097	-0.099
0.292	-0.973	-0.978	-0.986	-0.979		-1.015	-0.900	-1.028	-1.012	-0.997
0.322	0.035	0.073	-0.012	-0.014	-0.021	-0.149	-0.138	-0.083	-0.086	0.210
0.352	0.047	0.082	0.121	0.040	0.035	0.069	0.065	0.051	0.080	0.020
0.381	0.196	0.184		0.168	0.165	0.155	0.163	0.169	0.180	0.146
0.392	0.137	0.108	0.118	0.102	0.093	0.092	0.094	0.091	0.097	0.069
0.402	-0.008	-0.021	-0.029	-0.033	-0.033	-0.008	-0.017	-0.027	-0.028	-0.019
0.447	0.046	0.038	-0.009	0.023	0.014		0.012	0.007	0.014	0.004
0.492	0.042	0.037	0.027	0.017	0.005	0.010	0.010	0.010	0.009	-0.004
0.529	0.031	0.030	0.019	0.004	-0.000	-0.000	-0.000	-0.008	-0.005	-0.008
0.592	0.024	0.019	0.014	-0.001	-0.011	-0.016	-0.016	-0.011	-0.015	-0.015
0.655	-0.346	-0.050	-0.170	-0.187	-0.200	-0.178	-0.098	-0.178	-0.183	-0.192
0.661	-0.368	-0.760	-0.346	-0.242	-0.214	-0.219		-0.219	-0.207	-0.203
0.687	-0.014	0.049	-0.030	-0.058	-0.053	-0.029	0.013	-0.046		-0.063

(f) α = 6.0°

0.223	0.975	0.965	0.917	0.830	0.780	0.586	0.607	0.652	0.718	0.776
0.226	0.309	0.293	0.230	0.101	0.002	-0.284	-0.271	-0.203	-0.078	0.016
0.238	0.422	0.415	0.370	0.310	0.269	0.200	0.175	0.221	0.243	0.279
0.251	0.377	0.338	0.308	0.245	0.206	0.139	0.175	0.151	0.172	0.189
0.269	0.238	0.209	0.177	0.117	0.076	0.024	0.034	0.037	0.033	0.055
0.286	-0.193	0.031	-0.026	-0.084	-0.110	-0.147	-0.147	-0.151	-0.105	-0.119
0.292	-0.951	-0.943	-0.960	-0.960		-1.008	-0.882	-1.011	-1.011	-0.979
0.322	0.060	0.153	-0.036	-0.049	-0.046	-0.202	-0.208	-0.146	-0.155	0.143
0.352	0.056	0.071	0.132	0.012	-0.001	0.087	0.088	0.050	0.093	0.054
0.381	0.228	0.205		0.144	0.153	0.151	0.164	0.168	0.165	0.137
0.392	0.158	0.124	0.125	0.096	0.080	0.092	0.091	0.083	0.088	0.063
0.402	0.008	-0.004	-0.023	-0.044	-0.047	0.001	-0.016	-0.037	-0.036	-0.030
0.447	0.064	0.053	0.034	0.015	0.003		0.011	0.001	-0.003	-0.010
0.492	0.062	0.053	0.030	0.008	-0.008	0.013	0.010	0.004	-0.005	-0.006
0.529	0.058	0.052	0.025	-0.008	-0.020	-0.000	-0.002	-0.015	-0.029	-0.029
0.592	0.059	0.047	-0.198	-0.020	-0.037	-0.014	-0.019	-0.027	-0.041	-0.043
0.655	-0.078	-0.017	-0.150	-0.198	-0.229	-0.163	-0.098	-0.181	-0.214	-0.219
0.661	-0.363	-0.737	-0.404	-0.372	-0.290	-0.216		-0.199	-0.237	-0.256
0.687	0.015	0.071	-0.026	-0.081	-0.077	-0.018	0.019	-0.044		-0.092

[redacted]

[redacted]

TABLE II.- PRESSURE COEFFICIENTS OVER THE BODY WITH CHORD FIN FOR $M_\infty = 0.90$ - Concluded

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	75	90	180	200	225	250	270
0.223	1.023	1.008	0.944	0.838	0.758	0.504	0.526	0.586	0.676	0.756
0.226	0.393	0.371	0.283	0.113	-0.017	-0.423	-0.403	-0.302	-0.130	0.003
0.238	0.460	0.437	0.377	0.291	0.238	0.158	0.136	0.173	0.201	0.246
0.251	0.411	0.368	0.317	0.227	0.173	0.098	0.134	0.107	0.131	0.155
0.269	0.272	0.237	0.186	0.095	0.035	-0.018	-0.009	-0.007	-0.023	0.010
0.286	0.043	0.050	-0.028	-0.114	-0.152	-0.183	-0.184	-0.194	-0.172	-0.160
0.292	-0.941	-0.939	-0.967	-0.975		-1.026	-0.896	-1.025	-1.028	-1.014
0.322	0.057	0.182	-0.081	-0.107	-0.096	-0.200	-0.214	-0.177	-0.195	-0.064
0.352	0.055	0.053	0.128	-0.023	-0.044	0.081	0.076	0.031	0.077	-0.081
0.381	0.234	0.206		0.106	0.122	0.135	0.147	0.153	0.133	0.110
0.392	0.171	0.131	0.118	0.071	0.049	0.081	0.076	0.060	0.051	0.026
0.402	0.018	-0.001	-0.029	-0.071	-0.081	-0.000	-0.024	-0.058	-0.073	-0.065
0.447	0.077	0.061	0.021	-0.013	-0.032		-0.003	-0.021	-0.037	-0.052
0.492	0.079	0.063	0.019	-0.028	-0.050	-0.000	-0.006	-0.022	-0.048	-0.051
0.529	0.078	0.063	-0.001	-0.047	-0.043	-0.013	-0.017	-0.036	-0.071	-0.050
0.592	0.075	0.056	0.004	-0.053	-0.077	-0.026	-0.040	-0.045	-0.079	-0.083
0.655	-0.063	-0.023	-0.164	-0.230	-0.265	-0.166	-0.138	-0.195	-0.248	-0.254
0.661	-0.317	-0.745	-0.429	-0.402	-0.320	-0.209		-0.198	-0.263	-0.289
0.687	0.020	0.071	-0.040	-0.113	-0.110	-0.025	0.005	-0.054		-0.128

(h) $\alpha = 10.0°$

0.223	1.073	1.055	0.974	0.843	0.745	0.432	0.458	0.529	0.640	0.738
0.226	0.487	0.455	0.345	0.136	-0.023	-0.522	-0.500	-0.385	-0.168	-0.015
0.238	0.507	0.477	0.401	0.280	0.212	0.133	0.111	0.139	0.165	0.220
0.251	0.458	0.410	0.336	0.218	0.144	0.073	0.108	0.073	0.097	0.128
0.269	0.317	0.277	0.207	0.098	0.036	-0.044	-0.034	-0.038	0.056	0.010
0.286	0.058	0.062	-0.019	-0.109	-0.150	-0.179	-0.179	-0.189	-0.169	-0.157
0.292	-0.911	-0.919	-0.953	-0.963		-1.015	-0.885	-1.014	-1.017	-1.019
0.322	0.059	0.177	-0.070	-0.093	-0.085	-0.207	-0.222	-0.182	-0.193	-0.016
0.352	0.055	0.061	0.129	-0.005	-0.021	0.083	0.084	0.042	0.087	0.023
0.381	0.226	0.202		0.137	0.094	0.134	0.138	0.144	0.102	0.085
0.392	0.197	0.149	0.115	0.051	0.022	0.084	0.072	0.049	0.008	-0.003
0.402	0.044	0.016	-0.024	-0.087	-0.105	0.009	-0.026	-0.062	-0.083	-0.088
0.447	0.101	0.080	0.025	-0.026	-0.048		-0.000	-0.025	-0.047	-0.065
0.492	0.097	0.077	0.024	-0.033	-0.056	0.004	-0.003	-0.019	-0.050	-0.054
0.529	0.090	0.073	0.019	-0.045	-0.063	-0.007	-0.018	-0.030	-0.067	-0.070
0.592	0.085	0.066	0.011	-0.048	-0.073	-0.022	-0.035	-0.040	-0.075	-0.079
0.655	-0.056	-0.016	-0.158	-0.223	-0.259	-0.163	-0.136	-0.192	-0.242	-0.246
0.661	-0.319	-0.736	-0.421	-0.429	-0.361	-0.207		-0.203	-0.266	-0.297
0.687	0.015	0.070	-0.008	-.097	-0.004	-0.024	-0.205	-0.052		-0.105

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TABLE III.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE ON FOR $M_\infty = 1.60$

(a) $\alpha = 4.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	0.739	0.751	0.771	0.823	0.865	0.986	0.987	0.960	0.903	0.870
0.226	-0.089	-0.085	-0.030	0.042	0.094	0.292	0.286	0.235	0.164	0.105
0.238	0.314	0.323	0.325	0.351	0.372	0.463	0.450	0.441	0.397	0.379
0.251	0.279	0.258	0.277	0.291	0.319	0.404	0.407	0.372	0.327	0.294
0.269	0.158	0.145	0.158	0.192	0.207	0.286	0.307	0.270	0.233	0.197
0.286	0.040	0.061	0.050	0.057	0.069	0.106	0.105	0.089	0.157	0.061
0.292	-0.639	-0.750	-0.747	-0.743		-0.720	-0.719	-0.720	-0.736	-0.739
0.322	-0.349	-0.221	-0.183	-0.394	-0.392	-0.183	-0.315	-0.361	-0.144	0.042
0.352	-0.275	-0.246	-0.309	-0.280	-0.273	-0.275	-0.242	-0.276	-0.369	-0.438
0.381	0.155	0.165		0.161	0.155	0.178	0.184	0.179	0.150	0.124
0.392	0.111	0.102	0.102	0.109	0.118	0.165	0.166	0.149	0.123	0.088
0.402	-0.056	-0.074	-0.067	-0.100	-0.108	-0.071	-0.088	-0.099	-0.085	-0.066
0.447	0.040	0.043	0.035	0.039	0.038		0.059	0.048	0.043	0.031
0.492	0.062	0.063	0.060	0.063	0.060	0.085	0.085	0.080	0.073	0.064
0.529	0.054	0.058	0.056	0.055	0.058	0.075	0.076	0.069	0.060	0.053
0.592	0.047	0.048	0.047	0.050	0.051	0.068	0.068	0.066	0.058	0.049
0.655	-0.067	0.008	-0.069	-0.072	-0.077	-0.006	0.106	-0.027	-0.055	-0.078
0.661	-0.252	-0.562	-0.215	-0.257	-0.254	-0.225		-0.220	-0.235	-0.248
0.687	-0.129	-0.056	-0.158	-0.196	-0.166	-0.125	-0.039	-0.147		-0.185

(b) $\alpha = 8.2°$

0.223	0.807	0.813	0.826	0.846	0.872	0.930	0.933	0.922	0.890	0.874
0.226	0.027	0.009	0.041	0.077	0.098	0.203	0.201	0.173	0.140	0.110
0.238	0.349	0.355	0.355	0.370	0.381	0.425	0.394	0.418	0.394	0.388
0.251	0.313	0.291	0.307	0.312	0.327	0.369	0.400	0.353	0.326	0.305
0.269	0.194	0.177	0.188	0.199	0.210	0.257	0.272	0.251	0.232	0.200
0.286	0.048	0.066	0.055	0.061	0.072	0.103	0.104	0.088	0.081	0.063
0.292	-0.636	-0.744	-0.742	-0.739		-0.722	-0.720	-0.841	-0.734	-0.735
0.322	-0.326	-0.180	-0.182	-0.393	-0.392	-0.236	-0.355	-0.363	-0.174	0.042
0.352	-0.278	-0.284	-0.312	-0.282	-0.275	-0.276	-0.265	-0.277	-0.332	-0.392
0.381	0.152	0.162		0.159	0.164	0.174	0.181	0.179	0.156	0.150
0.392	0.123	0.113	0.118	0.127	0.131	0.152	0.155	0.147	0.132	0.107
0.402	-0.070	-0.087	-0.074	-0.085	-0.087	-0.073	-0.085	-0.086	-0.070	-0.068
0.447	0.036	0.038	0.033	0.040	0.039		0.047	0.042	0.042	0.032
0.492	0.062	0.063	0.060	0.064	0.060	0.075	0.075	0.073	0.069	0.061
0.529	0.054	0.057	0.055	0.055	0.056	0.068	0.070	0.064	0.058	0.052
0.592	0.049	0.050	0.048	0.052	0.053	0.067	0.067	0.066	0.058	0.050
0.655	-0.066	0.039	-0.068	-0.071	-0.076	-0.006	0.110	-0.026	-0.054	-0.078
0.661	-0.250	-0.558	-0.213	-0.225	-0.228	-0.224		-0.218	-0.233	-0.220
0.687	-0.127	-0.055	-0.157	-0.194	-0.164	-0.124	-0.038	-0.146		-0.183

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TABLE III.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_6 FOR M_∞=1.00 - Continued

(c) α = -0.1°

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.223	0.868	0.871	0.869	0.868	0.871	0.868	0.876	0.880	0.871	0.873
0.226	0.104	0.101	0.110	0.104	0.095	0.104	0.105	0.100	0.105	0.107
0.238	0.383	0.385	0.379	0.382	0.382	0.386	0.356	0.392	0.384	0.389
0.251	0.346	0.322	0.329	0.324	0.326	0.328	0.362	0.326	0.317	0.308
0.269	0.226	0.208	0.212	0.210	0.211	0.221	0.233	0.224	0.219	0.200
0.286	0.069	0.087	0.070	0.067	0.072	0.074	0.076	0.068	0.074	0.064
0.292	-0.637	-0.740	-0.739	-0.738		-0.738	-0.735	-0.855	-0.740	-0.738
0.322	-0.278	-0.117	-0.161	-0.367	-0.296	-0.298	-0.404	-0.387	-0.216	0.035
0.352	-0.275	-0.317	-0.289	-0.267	-0.276	-0.288	-0.279	-0.273	-0.297	-0.372
0.381	0.171	0.168		0.166	0.165	0.164	0.170	0.170	0.159	0.162
0.392	0.134	0.120	0.132	0.133	0.133	0.135	0.138	0.136	0.133	0.113
0.402	-0.081	-0.094	-0.089	-0.080	-0.078	-0.069	-0.077	-0.075	-0.073	-0.067
0.447	0.045	0.045	0.045	0.049	0.046		0.043	0.040	0.043	0.036
0.492	0.066	0.065	0.064	0.066	0.060	0.065	0.065	0.067	0.067	0.064
0.529	0.058	0.061	0.060	0.058	0.059	0.058	0.060	0.059	0.058	0.056
0.592	0.055	0.054	0.055	0.055	0.053	0.052	0.053	0.056	0.055	0.052
0.655	-0.037	0.053	-0.044	-0.065	-0.082	-0.047	0.066	-0.051	-0.063	-0.074
0.661	-0.242	-0.535	-0.230	-0.229	-0.226	-0.245		-0.216	-0.234	-0.221
0.687	-0.129	-0.048	-0.148	-0.199	-0.175	-0.133	-0.064	-0.163		-0.175

(d) α = 1.9°

0.223	0.927	0.926	0.909	0.883	0.869	0.807	0.819	0.835	0.851	0.869
0.226	0.195	0.189	0.171	0.128	0.092	0.023	0.006	0.029	0.074	0.105
0.238	0.417	0.418	0.402	0.388	0.378	0.348	0.321	0.364	0.369	0.385
0.251	0.380	0.352	0.351	0.331	0.321	0.293	0.328	0.300	0.302	0.303
0.269	0.258	0.237	0.232	0.219	0.201	0.187	0.197	0.197	0.185	0.187
0.286	0.107	0.123	0.091	0.067	0.059	0.033	0.036	0.034	0.052	0.051
0.292	-0.637	-0.729	-0.732	-0.740		-0.752	-0.750	-0.870	-0.751	-0.744
0.322	-0.207	-0.040	-0.136	-0.330	-0.397	-0.349	-0.442	-0.404	-0.256	0.037
0.352	-0.260	-0.330	-0.273	-0.262	-0.284	-0.274	-0.270	-0.274	-0.284	-0.404
0.381	0.169	0.157		0.155	0.158	0.149	0.155	0.157	0.170	0.149
0.392	0.145	0.126	0.140	0.131	0.125	0.119	0.121	0.121	0.134	0.104
0.402	-0.087	-0.103	-0.123	-0.095	-0.096	-0.073	-0.080	-0.086	-0.107	-0.078
0.447	0.038	0.035	0.034	0.032	0.026		0.023	0.020	0.029	0.018
0.492	0.071	0.068	0.064	0.059	0.049	0.053	0.053	0.053	0.054	0.050
0.529	0.070	0.070	0.061	0.049	0.044	0.044	0.046	0.042	0.041	0.041
0.592	0.074	0.069	0.060	0.050	0.042	0.038	0.038	0.041	0.038	0.039
0.655	0.007	0.088	-0.032	-0.063	-0.086	-0.089	0.012	-0.074	-0.085	-0.080
0.661	-0.222	-0.548	-0.236	-0.248	-0.255	-0.266		-0.223	-0.245	-0.247
0.687	-0.118	-0.033	-0.142	-0.210	-0.193	-0.137	-0.080	-0.171		-0.180

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TABLE III.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE NO. FOR $M_\infty = 1.60$ - Continued

(e) $\alpha = 3.9°$

Axial position x/l	Radial position, θ, deg									
	0	20	45	70	90	180	290	315	340	360
0.223	0.985	0.980	0.949	0.899	0.866	0.742	0.757	0.788	0.827	0.863
0.226	0.290	0.277	0.237	0.153	0.087	-0.095	-0.084	-0.041	0.039	0.103
0.238	0.458	0.455	0.426	0.392	0.368	0.316	0.289	0.336	0.352	0.378
0.251	0.419	0.387	0.374	0.334	0.311	0.261	0.298	0.272	0.285	0.294
0.269	0.296	0.272	0.256	0.222	0.201	0.157	0.165	0.171	0.182	0.187
0.286	0.117	0.131	0.096	0.069	0.059	0.031	0.034	0.033	0.052	0.052
0.292	-0.631	-0.720	-0.727	-0.737		-0.751	-0.749	-0.869	-0.749	-0.742
0.322	-0.201	-0.035	-0.133	-0.325	-0.394	-0.350	-0.443	-0.403	-0.256	-0.000
0.352	-0.258	-0.329	-0.271	-0.260	-0.282	-0.273	-0.268	-0.272	-0.281	-0.452
0.381	0.171	0.158		0.157	0.152	0.136	0.142	0.147	0.171	0.120
0.392	0.162	0.135	0.145	0.131	0.117	0.110	0.110	0.110	0.124	0.086
0.402	-0.091	-0.091	-0.129	-0.129	-0.113	-0.058	-0.063	-0.093	-0.121	-0.075
0.447	0.054	0.047	0.043	0.038	0.028		0.029	0.024	0.029	0.016
0.492	0.080	0.076	0.067	0.058	0.047	0.054	0.054	0.054	0.052	0.050
0.529	0.077	0.076	0.064	0.050	0.045	0.046	0.047	0.043	0.041	0.041
0.592	0.079	0.073	0.062	0.052	0.043	0.040	0.039	0.043	0.039	0.040
0.655	0.012	0.092	-0.028	-0.061	-0.084	-0.089	0.011	-0.073	-0.083	-0.077
0.661	-0.218	-0.547	-0.234	-0.245	-0.253	-0.263		-0.220	-0.244	-0.245
0.687	-0.115	-0.031	-0.139	-0.208	-0.190	-0.135	-0.077	-0.169		-0.177

(f) $\alpha = 6.0°$

0.223	1.038	1.029	0.984	0.910	0.857	0.677	0.693	0.737	0.799	0.852
0.226	0.382	0.363	0.300	0.179	0.084	-0.201	-0.187	-0.117	0.019	0.096
0.238	0.500	0.492	0.448	0.391	0.354	0.285	0.259	0.304	0.327	0.361
0.251	0.458	0.423	0.393	0.333	0.295	0.230	0.358	0.243	0.262	0.279
0.269	0.335	0.307	0.277	0.216	0.173	0.126	0.134	0.140	0.128	0.152
0.286	0.162	0.172	0.112	0.051	0.025	-0.006	-0.005	-0.009	0.039	0.018
0.292	-0.625	-0.701	-0.717	-0.742		-0.756	-0.760	-0.887	-0.766	-0.754
0.322	-0.145	0.072	-0.104	-0.278	-0.375	-0.366	-0.468	-0.412	-0.318	-0.036
0.352	-0.218	-0.329	-0.256	-0.280	-0.299	-0.183	-0.198	-0.258	-0.251	-0.501
0.381	0.162	0.134		0.135	0.138	0.124	0.127	0.134	0.150	0.100
0.392	0.177	0.143	0.144	0.122	0.099	0.101	0.096	0.094	0.108	0.081
0.402	-0.087	-0.083	-0.109	-0.175	-0.154	-0.043	-0.055	-0.109	-0.099	-0.087
0.447	0.076	0.065	0.041	0.010	0.010		0.028	0.018	0.005	-0.004
0.492	0.106	0.095	0.066	0.036	0.019	0.054	0.048	0.040	0.025	0.036
0.529	0.108	0.099	0.066	0.025	0.014	0.043	0.039	0.025	0.009	0.008
0.592	0.111	0.098	0.063	0.025	0.008	0.033	0.026	0.012	0.005	0.004
0.655	0.053	0.108	-0.022	-0.276	-0.114	-0.083	-0.026	-0.111	-0.107	-0.103
0.661	-0.191	-0.543	-0.254	-0.268	-0.298	-0.304		-0.245	-0.272	-0.289
0.687	-0.075	-0.003	-0.122	-0.228	-0.230	-0.130	-0.091	-0.182		-0.213

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TABLE III.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE NO. FOR $M_\infty = 1.60$ - Concluded

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	250	270
0.223	1.086	1.072	1.011	0.910	0.839	0.600	0.620	0.676	0.760	0.834
0.226	0.466	0.438	0.351	0.193	0.066	-0.312	-0.294	-0.207	-0.040	0.078
0.238	0.534	0.514	0.455	0.372	0.321	0.246	0.219	0.259	0.284	0.329
0.251	0.492	0.449	0.400	0.314	0.261	0.189	0.225	0.197	0.222	0.245
0.269	0.366	0.333	0.284	0.194	0.137	0.084	0.093	0.097	0.080	0.113
0.286	0.177	0.182	0.106	0.020	-0.015	-0.037	-0.037	-0.049	-0.035	-0.023
0.292	-0.638	-0.704	-0.730	-0.767		-0.775	-0.784	-0.783	-0.799	-0.785
0.322	-0.096	0.118	-0.108	-0.284	-0.379	-0.415	-0.485	-0.431	-0.371	-0.108
0.352	-0.208	-0.337	-0.246	-0.337	-0.311	-0.160	-0.178	-0.254	-0.261	-0.532
0.381	0.134	0.111		0.095	0.109	0.110	0.102	0.105	0.111	0.063
0.392	0.182	0.141	0.124	0.090	0.067	0.094	0.079	0.070	0.066	0.040
0.402	-0.072	-0.074	-0.095	-0.191	-0.216	-0.028	-0.038	-0.093	-0.111	-0.115
0.447	0.093	0.076	0.035	0.001	-0.015		0.019	0.006	-0.015	-0.033
0.492	0.116	0.098	0.054	0.008	-0.011	0.043	0.031	0.017	-0.005	-0.013
0.529	0.116	0.101	0.051	-0.208	-0.023	0.027	0.017	0.006	-0.026	-0.031
0.592	0.119	0.101	0.048	-0.008	-0.030	0.018	0.015	-0.001	-0.031	-0.036
0.655	0.059	0.097	-0.040	-0.112	-0.121	-0.092	-0.056	-0.126	-0.144	-0.143
0.661	-0.190	-0.555	-0.283	-0.334	-0.343	-0.315		-0.269	-0.344	-0.333
0.687	-0.059	-0.000	-0.127	-0.246	-0.266	-0.129	-0.103	-0.201		-0.256

(h) $\alpha = 10.0°$

0.223	1.130	1.111	1.036	0.912	0.822	0.533	0.556	0.621	0.725	0.817
0.226	0.551	0.518	0.408	0.216	0.060	-0.384	-0.368	-0.275	-0.074	0.069
0.238	0.576	0.547	0.474	0.357	0.291	0.220	0.190	0.222	0.247	0.299
0.251	0.533	0.486	0.416	0.302	0.230	0.162	0.196	0.162	0.186	0.215
0.269	0.404	0.367	0.299	0.334	0.228	0.057	0.067	0.063	0.117	0.182
0.286	0.319	0.169	0.181	0.022	-0.049	-0.081	0.065	-0.104	-0.082	-0.059
0.292	-0.681	-0.690	-0.721	-0.760		-0.765	-0.778	-0.792	-0.791	-0.778
0.322	-0.036	0.125	-0.104	-0.281	-0.377	-0.407	-0.477	-0.424	-0.347	-0.175
0.352	-0.374	-0.329	-0.445	-0.332	-0.324	-0.297	-0.168	-0.463	-0.253	-0.478
0.381	0.111	0.183	-0.008	0.160	0.084	0.110	0.080	0.080	0.062	0.032
0.392	0.192	0.144	0.103	0.058	0.032	0.100	0.064	0.049	0.003	0.004
0.402	-0.046	-0.059	-0.081	-0.176	-0.236	-0.009	-0.030	-0.064	-0.114	-0.143
0.447	0.117	0.095	0.042	-0.008	-0.025		0.022	0.007	-0.017	-0.040
0.492	0.132	0.111	0.059	0.004	-0.015	0.050	0.034	0.020	-0.005	-0.014
0.529	0.126	0.110	0.056	-0.008	-0.024	0.032	0.018	0.010	-0.025	-0.031
0.592	0.124	0.105	0.050	-0.008	-0.030	0.020	0.001	-0.000	-0.031	-0.036
0.655	0.062	0.099	-0.038	-0.111	-0.153	-0.067	-0.054	-0.139	-0.142	-0.141
0.661	-0.185	-0.551	-0.280	-0.333	-0.341	-0.311		-0.266	-0.342	-0.331
0.687	-0.055	0.002	-0.124	-0.243	-0.263	-0.126	-0.100	-0.197		-0.254

[redacted]

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TABLE IV.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE NO. FOR M_∞ = 1.10

(a) α = -4.1°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	0.819	0.830	0.854	0.895	0.937	1.053	1.053	1.028	0.975	0.941
0.226	-0.008	-0.017	0.034	0.099	0.148	0.344	0.338	0.287	0.218	0.164
0.238	0.385	0.390	0.393	0.417	0.439	0.527	0.495	0.506	0.465	0.447
0.251	0.357	0.338	0.354	0.368	0.395	0.480	0.504	0.447	0.403	0.372
0.269	0.257	0.242	0.254	0.272	0.292	0.377	0.396	0.361	0.327	0.283
0.286	0.150	0.166	0.159	0.170	0.186	0.248	0.246	0.222	0.199	0.177
0.292	-0.560	-0.557	-0.557	-0.553		-0.521	-0.521	-0.530	-0.544	-0.548
0.322	-0.246	-0.107	-0.088	-0.292	-0.275	-0.114	-0.215	-0.232	-0.051	0.109
0.352	-0.190	-0.196	-0.252	-0.207	-0.179	-0.156	-0.142	-0.171	-0.267	-0.331
0.381	0.159	0.167		0.158	0.148	0.120	0.135	0.151	0.139	0.115
0.392	0.172	0.169	0.163	0.177	0.187	0.221	0.227	0.212	0.183	0.154
0.402	0.005	0.170	-0.005	-0.029	-0.029	-0.003	0.228	-0.022	-0.021	-0.000
0.447	0.021	0.024	0.015	0.020	0.020		0.050	0.037	0.029	0.013
0.492	0.019	0.019	0.015	0.013	0.008	0.045	0.044	0.035	0.022	0.006
0.529	0.027	0.031	0.026	0.023	0.026	0.057	0.055	0.042	0.027	0.017
0.592	0.025	0.025	0.019	0.018	0.019	0.060	0.058	0.049	0.033	0.022
0.655	-0.005	0.080	-0.014	-0.011	-0.016	0.099	0.211	0.063	0.020	-0.007
0.661	-0.151	-0.441	-0.126	-0.158	-0.157	-0.095		-0.109	-0.140	-0.146
0.687	-0.077	-0.011	-0.112	-0.121	-0.089	-0.078	0.022	-0.110		-0.130

(b) α = -2.2°

0.223	0.880	0.886	0.897	0.916	0.941	0.999	1.001	0.990	0.961	0.944
0.226	0.069	0.071	0.098	0.129	0.150	0.252	0.249	0.221	0.192	0.162
0.238	0.416	0.421	0.423	0.437	0.449	0.491	0.460	0.484	0.461	0.457
0.251	0.389	0.368	0.382	0.389	0.404	0.443	0.470	0.427	0.403	0.381
0.269	0.286	0.270	0.280	0.290	0.301	0.344	0.359	0.340	0.322	0.291
0.286	0.170	0.187	0.177	0.182	0.193	0.223	0.223	0.209	0.201	0.185
0.292	-0.558	-0.555	-0.552	-0.549		-0.535	-0.534	-0.534	-0.545	-0.546
0.322	-0.225	-0.072	-0.077	-0.280	-0.274	-0.155	-0.245	-0.251	-0.072	0.124
0.352	-0.186	-0.216	-0.232	-0.189	-0.182	-0.171	-0.164	-0.177	-0.235	-0.294
0.381	0.164	0.167		0.166	0.164	0.148	0.163	0.168	0.148	0.128
0.392	0.186	0.181	0.181	0.191	0.196	0.212	0.218	0.210	0.193	0.174
0.402	-0.003	-0.025	-0.007	-0.014	-0.013	-0.001	-0.010	-0.013	-0.000	0.004
0.447	0.026	0.027	0.022	0.030	0.029		0.039	0.034	0.033	0.023
0.492	0.022	0.023	0.023	0.026	0.021	0.034	0.034	0.032	0.028	0.021
0.529	0.034	0.038	0.036	0.036	0.038	0.048	0.048	0.041	0.034	0.028
0.592	0.028	0.028	0.025	0.027	0.027	0.044	0.046	0.046	0.040	0.033
0.655	0.008	0.115	0.004	-0.007	-0.014	0.068	0.185	0.046	0.009	-0.016
0.661	-0.146	-0.419	-0.117	-0.136	-0.140	-0.113		-0.113	-0.139	-0.140
0.687	-0.083	-0.004	-0.112	-0.119	-0.084	-0.087	0.008	-0.117		-0.109

[redacted]

TABLE IV.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_5 FOR $M_\infty = 1.10$ - Continued

(c) $\alpha = -0.1°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	0.934	0.936	0.936	0.933	0.938	0.937	0.943	0.946	0.939	0.941
0.226	0.152	0.151	0.156	0.151	0.141	0.150	0.152	0.148	0.155	0.157
0.238	0.445	0.448	0.440	0.443	0.445	0.448	0.419	0.453	0.448	0.452
0.251	0.417	0.393	0.402	0.397	0.399	0.400	0.431	0.398	0.389	0.379
0.269	0.313	0.294	0.297	0.298	0.299	0.306	0.318	0.311	0.307	0.288
0.286	0.187	0.203	0.187	0.186	0.190	0.195	0.196	0.189	0.192	0.183
0.292	-0.558	-0.555	-0.550	-0.553		-0.552	-0.550	-0.555	-0.554	-0.552
0.322	-0.203	-0.037	-0.069	-0.273	-0.280	-0.196	-0.278	-0.273	-0.100	0.120
0.352	-0.183	-0.237	-0.221	-0.178	-0.190	-0.191	-0.188	-0.182	-0.223	-0.296
0.381	0.156	0.149		0.160	0.163	0.157	0.168	0.169	0.147	0.146
0.392	0.195	0.185	0.193	0.194	0.194	0.194	0.007	0.196	0.191	0.177
0.402	-0.015	-0.032	-0.029	-0.016	-0.014	-0.005	-0.012	-0.011	-0.009	-0.006
0.447	0.028	0.027	0.027	0.031	0.027		0.025	0.023	0.028	0.021
0.492	0.021	0.022	0.023	0.027	0.022	0.023	0.023	0.024	0.023	0.019
0.529	0.034	0.037	0.035	0.033	0.033	0.034	0.036	0.034	0.032	0.030
0.592	0.031	0.030	0.030	0.031	0.030	0.032	0.034	0.038	0.037	0.034
0.655	0.026	0.123	0.021	-0.002	-0.019	0.030	0.146	0.019	-0.007	-0.014
0.661	-0.143	-0.400	-0.122	-0.135	-0.141	-0.137		-0.121	-0.141	-0.157
0.687	-0.095	-0.010	-0.116	-0.129	-0.093	-0.094	-0.011	-0.127		-0.102

(d) $\alpha = 1.9°$

0.223	0.992	0.990	0.976	0.951	0.939	0.881	0.891	0.907	0.922	0.939
0.226	0.246	0.239	0.231	0.179	0.145	0.063	0.072	0.087	0.130	0.160
0.238	0.483	0.483	0.467	0.455	0.444	0.417	0.390	0.430	0.436	0.453
0.251	0.454	0.428	0.427	0.407	0.397	0.370	0.404	0.377	0.380	0.380
0.269	0.349	0.328	0.324	0.312	0.294	0.280	0.288	0.289	0.278	0.280
0.286	0.230	0.244	0.214	0.262	0.181	0.157	0.159	0.158	0.264	0.175
0.292	-0.538	-0.538	0.256	-0.548		-0.558	-0.556	-0.548	-0.558	-0.552
0.322	-0.130	0.052	-0.038	-0.238	-0.279	-0.241	-0.314	-0.278	-0.109	0.115
0.352	-0.152	-0.244	-0.202	-0.157	-0.184	-0.199	-0.197	-0.175	-0.212	-0.311
0.381	0.118	0.102		0.133	0.162	0.163	0.170	0.168	0.167	0.136
0.392	0.210	0.193	0.209	0.199	0.194	0.184	0.188	0.188	0.203	0.173
0.402	-0.014	-0.021	-0.053	-0.019	-0.015	0.005	-0.000	-0.006	-0.046	0.002
0.447	0.048	0.043	0.041	0.037	0.029		0.023	0.019	0.029	0.017
0.492	0.042	0.040	0.034	0.028	0.017	0.024	0.022	0.019	0.018	0.015
0.529	0.054	0.054	0.044	0.031	0.026	0.026	0.027	0.022	0.020	0.020
0.592	0.052	0.047	0.037	0.028	0.020	0.024	0.024	0.027	0.023	0.022
0.655	0.088	0.173	0.049	0.011	-0.014	-0.015	0.080	-0.003	-0.020	-0.018
0.661	-0.105	-0.401	-0.121	-0.140	-0.151	-0.163		-0.132	-0.160	-0.158
0.687	-0.085	0.015	-0.102	-0.147	-0.117	-0.087	-0.030	-0.129		-0.102

[redacted]

TABLE IV.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE No. FOR M_∞ = 1.10 - Continued

(e) α = [illegible]°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	288	315	330	360
0.223	1.048	1.041	1.012	0.965	0.934	0.820	0.833	0.861	0.899	0.934
0.226	0.340	0.329	0.295	0.206	0.142	-0.028	-0.021	0.017	0.097	0.156
0.238	0.521	0.518	0.489	0.457	0.434	0.383	0.355	0.401	0.416	0.443
0.251	0.491	0.461	0.446	0.477	0.386	0.337	0.368	0.348	0.434	0.367
0.269	0.386	0.359	0.343	0.381	0.287	0.249	0.256	0.262	0.347	0.270
0.286	0.245	0.258	0.220	0.260	0.169	0.141	0.143	0.141	0.248	0.161
0.292	-0.531	-0.531	-0.467	-0.476		-0.563	-0.565	-0.484	-0.479	-0.476
0.322	-0.055	0.127	0.032	-0.222	-0.203	-0.166	-0.240	-0.217	-0.173	0.159
0.352	-0.137	-0.247	-0.198	-0.166	-0.110	-0.194	-0.194	-0.186	-0.210	-0.277
0.381	0.083	0.082		0.208	0.152	0.158	0.164	0.239	0.246	0.113
0.392	0.221	0.194	0.279	0.270	0.184	0.171	0.172	0.252	0.269	0.153
0.402	-0.018	-0.017	-0.048	-0.042	-0.031	0.009	0.001	-0.019	-0.045	-0.002
0.447	0.054	0.046	0.040	0.032	0.022		0.017	0.002	0.021	0.010
0.492	0.047	0.044	0.033	0.022	0.010	0.023	0.022	0.016	0.012	0.009
0.529	0.060	0.058	0.042	0.021	0.011	0.021	0.022	0.017	0.013	0.004
0.592	0.066	0.058	0.041	0.022	0.013	0.021	0.018	0.026	0.006	0.005
0.655	0.110	0.185	0.057	0.010	-0.025	-0.023	0.061	-0.009	-0.029	-0.027
0.661	-0.091	-0.402	-0.127	-0.157	-0.164	-0.179		-0.135	-0.161	-0.172
0.687	-0.077	0.023	-0.098	-0.164	-0.143	-0.081	-0.033	-0.131		-0.119

(f) α = 6.0°

0.223	1.100	1.089	1.047	0.974	0.926	0.758	0.774	0.815	0.872	0.924
0.226	0.430	0.412	0.348	0.231	0.140	-0.110	-0.102	-0.043	0.065	0.153
0.238	0.560	0.553	0.511	0.454	0.418	0.354	0.325	0.371	0.391	0.425
0.251	0.528	0.495	0.463	0.405	0.368	0.305	0.318	0.318	0.337	0.317
0.269	0.421	0.351	0.421	0.307	0.278	0.231	0.362	0.221	0.248	0.277
0.286	0.244	0.259	0.262	0.271	0.224	0.178	0.158	0.135	0.137	0.131
0.292	-0.519	-0.522	-0.521	-0.522		-0.553	-0.563	-0.562	-0.566	-0.556
0.322	-0.084	0.101	-0.026	-0.222	-0.280	-0.258	-0.327	-0.297	-0.162	0.045
0.352	-0.180	-0.243	-0.131	-0.246	-0.194	-0.168	-0.183	-0.190	-0.190	-0.402
0.381	0.083	0.083		0.080	0.129	0.150	0.153	0.154	0.153	0.091
0.392	0.228	0.192	0.199	0.187	0.166	0.158	0.156	0.158	0.172	0.138
0.402	-0.021	-0.020	-0.034	-0.084	-0.062	0.015	-0.186	-0.044	-0.023	-0.018
0.447	0.067	0.056	0.033	0.016	0.002		0.011	0.003	-0.001	-0.006
0.492	0.056	0.047	0.025	0.006	-0.008	0.023	0.014	0.005	-0.006	-0.000
0.529	0.071	0.066	0.044	0.016	0.006	0.019	0.005	0.008	-0.001	-0.001
0.592	0.071	0.060	0.036	0.011	-0.002	0.026	0.016	0.010	0.003	0.001
0.655	0.119	0.188	0.059	0.009	-0.028	-0.022	0.056	-0.033	-0.033	-0.030
0.661	-0.085	-0.399	-0.127	-0.160	-0.166	-0.182		-0.138	-0.184	-0.176
0.687	-0.074	0.026	-0.094	-0.166	-0.146	-0.078	-0.032	-0.131		-0.124

[redacted]

[redacted]

TABLE IV.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE NO. FOR $M_\infty = 1.10$ - Concluded

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.223	1.150	1.136	1.078	0.984	0.918	0.698	0.716	0.769	0.845	0.915
0.226	0.523	0.499	0.414	0.262	0.144	-0.179	-0.163	-0.099	0.053	0.155
0.238	0.604	0.587	0.528	0.446	0.400	0.333	0.303	0.340	0.365	0.405
0.251	0.570	0.530	0.483	0.400	0.349	0.282	0.312	0.288	0.310	0.332
0.269	0.463	0.429	0.381	0.300	0.247	0.200	0.207	0.208	0.194	0.224
0.286	0.305	0.309	0.236	0.151	0.112	0.090	0.120	0.064	0.085	0.105
0.292	-0.496	-0.503	-0.527	-0.562		-0.560	-0.573	-0.572	-0.595	-0.579
0.322	0.029	0.210	-0.001	-0.189	-0.286	-0.281	-0.344	-0.306	-0.244	-0.005
0.352	-0.091	-0.239	-0.153	-0.235	-0.204	-0.145	-0.154	-0.192	-0.163	-0.418
0.381	0.040	0.071		0.106	0.102	0.152	0.145	0.146	0.124	0.070
0.392	0.238	0.194	0.181	0.167	0.147	0.155	0.142	0.141	0.130	0.109
0.402	-0.018	-0.029	-0.031	-0.104	-0.108	0.017	0.005	-0.078	-0.053	-0.066
0.447	0.086	0.070	0.031	-0.002	-0.022		0.006	-0.005	-0.021	-0.035
0.492	0.083	0.068	0.027	-0.018	-0.038	0.020	0.009	-0.004	-0.034	-0.041
0.529	0.102	0.088	0.040	-0.018	-0.035	0.018	0.009	-0.001	-0.032	-0.036
0.592	0.113	0.093	0.040	-0.015	-0.036	0.021		-0.004	-0.014	-0.043
0.655	0.160	0.197	0.058	-0.016	-0.058	0.028	0.030	-0.055	-0.061	-0.062
0.661	-0.057	-0.394	-0.146	-0.177	-0.209	-0.179		-0.151	-0.219	-0.214
0.687	-0.037	0.046	-0.079	-0.197	-0.199	-0.055	-0.030	-0.129		-0.174

(h) $\alpha = 10.0°$

0.223	1.192	1.173	1.100	0.982	0.898	0.627	0.648	0.710	0.807	0.893
0.226	0.604	0.574	0.468	0.280	0.137	-0.252	-0.238	-0.161	0.009	0.142
0.238	0.640	0.612	0.541	0.425	0.361	0.308	0.270	0.297	0.319	0.366
0.251	0.604	0.559	0.491	0.379	0.311	0.246	0.275	0.243	0.266	0.294
0.269	0.495	0.455	0.389	0.293	0.243	0.165	0.174	0.167	0.194	0.222
0.286	0.289	0.290	0.226	0.152	0.117	0.099	0.098	0.085	0.104	0.118
0.292	-0.510	-0.514	-0.540	-0.571		-0.572	-0.581	-0.605	-0.597	-0.586
0.322	0.021	0.119	-0.026	-0.218	-0.291	-0.277	-0.344	-0.311	-0.192	-0.085
0.352	-0.137	-0.254	-0.200	-0.182	-0.195	-0.198	-0.199	-0.194	-0.214	-0.393
0.381	0.081	0.078		0.090	0.048	0.143	0.118	0.118	0.056	0.039
0.392	0.237	0.188	0.142	0.121	0.105	0.150	0.114	0.106	0.048	0.060
0.402	-0.022	-0.055	-0.054	-0.125	-0.161	0.006	-0.008	-0.096	-0.078	-0.128
0.447	0.094	0.071	0.020	-0.031	-0.052		-0.010	-0.026	-0.047	-0.063
0.492	0.082	0.064	0.015	-0.037	-0.057	0.008	-0.003	-0.019	-0.050	-0.056
0.529	0.090	0.078	0.035	-0.019	-0.033	0.010	0.001	-0.008	-0.032	-0.033
0.592	0.087	0.071	0.031	-0.009	-0.026	0.011	0.002	-0.003	-0.020	-0.022
0.655	0.124	0.184	0.052	-0.004	-0.042	-0.017	0.044	-0.044	-0.045	-0.040
0.661	-0.088	-0.405	-0.138	-0.175	-0.179	-0.188		-0.142	-0.190	-0.181
0.687	-0.081	0.016	-0.106	-0.173	-0.150	-0.086	-0.038	-0.138		-0.123

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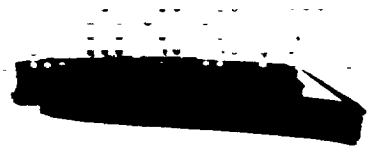

TABLE V.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_6 FOR M_∞=1.90

(a) α = -4.2

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	0.877	0.887	0.910	0.946	0.986	1.086	1.085	1.062	1.014	0.984
0.226	0.029	0.027	0.066	0.111	0.149	0.295	0.290	0.251	0.201	0.156
0.238	0.386	0.389	0.386	0.395	0.416	0.498	0.468	0.475	0.437	0.420
0.251	0.354	0.336	0.353	0.367	0.395	0.477	0.501	0.443	0.398	0.366
0.269	0.304	0.290	0.299	0.329	0.341	0.405	0.423	0.390	0.362	0.330
0.286	0.190	0.214	0.216	0.232	0.248	0.294	0.292	0.275	0.261	0.242
0.292	-0.427	-0.425	-0.421	-0.415		-0.393	-0.393	-0.468	-0.409	-0.412
0.322	-0.175	-0.021	-0.025	-0.218	-0.210	-0.130	-0.184	-0.191	-0.032	0.150
0.352	-0.142	-0.196	-0.200	-0.142	-0.135	-0.129	-0.113	-0.126	-0.198	-0.277
0.381	0.092	0.056		0.074	0.033	-0.000	-0.024	0.009	0.041	0.024
0.392	0.165	0.162	0.154	0.170	0.179	0.200	0.213	0.203	0.169	0.141
0.402	0.021	-0.042	0.010	-0.005	-0.000	0.020	0.017	0.009	-0.000	0.020
0.447	0.021	0.024	0.018	0.024	0.024		0.047	0.035	0.029	0.011
0.492	0.004	0.018	0.015	0.029	0.014	0.042	0.041	0.035	0.025	0.004
0.529	0.024	0.028	0.024	0.024	0.027	0.045	0.029	0.039	0.031	0.025
0.592	0.028	0.030	0.028	0.030	0.031	0.046	0.046	0.045	0.037	0.029
0.655	0.034	0.138	0.025	0.010	0.002	0.090	0.205	0.065	0.023	0.005
0.661	-0.096	-0.039	-0.081	-0.114	-0.115	-0.059		-0.065	-0.108	-0.112
0.687	-0.076	0.006	-0.111	-0.092	-0.064	-0.086	0.023	-0.115		-0.079

(b) α = -2.2°

0.223	0.930	0.936	0.949	0.967	0.990	1.037	1.038	1.029	0.999	0.986
0.226	0.090	0.089	0.114	0.133	0.149	0.224	0.222	0.200	0.178	0.153
0.238	0.407	0.411	0.408	0.419	0.428	0.463	0.436	0.458	0.436	0.433
0.251	0.387	0.366	0.380	0.386	0.403	0.441	0.468	0.423	0.398	0.377
0.269	0.332	0.316	0.325	0.338	0.346	0.380	0.395	0.375	0.359	0.334
0.286	0.217	0.237	0.233	0.241	0.251	0.276	0.275	0.263	0.258	0.245
0.292	-0.418	-0.416	-0.412	-0.409		-0.400	-0.398	-0.471	-0.407	-0.407
0.322	-0.162	0.007	-0.016	-0.208	-0.205	-0.147	-0.199	-0.199	-0.046	0.156
0.352	-0.131	-0.204	-0.189	-0.126	-0.136	-0.137	-0.129	-0.128	-0.187	-0.261
0.381	0.068	0.030		0.066	0.066	0.050	0.061	0.026	0.177	0.029
0.392	0.170	0.183	0.189	0.196	0.207	0.200	0.180	0.162	0.183	0.163
0.402	0.012	0.007	0.011	0.021	0.017	0.015	0.020	0.028	0.019	0.026
0.447	0.019	0.033	0.032	0.006	0.041		0.020	0.018	0.020	0.022
0.492	0.015	0.024	0.008	0.036	0.036	0.034	0.019	0.035	0.019	0.014
0.529	0.029	0.029	0.031	0.040	0.042	0.038	0.034	0.030	0.033	0.034
0.592	0.034	0.036	0.035	0.040	0.040	0.042	0.039	0.034	0.050	0.153
0.655	0.041	0.016	0.003	0.006	0.005	0.054	0.020	0.012	0.021	0.005
0.661	-0.072	-0.107	-0.110	-0.075	0.005	-0.068		-0.113	-0.081	0.009
0.687	-0.110	-0.092	-0.061	-0.081	-0.062	-0.116	0.003	-0.069		-0.072

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TABLE V.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_4 FOR $M_\infty = 1.20$ - Continued

(c) $\alpha = -0.1°$

Axial position x/l	Radial position, θ, deg									
	0	0	45	72	90	180	200	225	252	270
0.223	0.982	0.986	0.987	0.984	0.991	0.988	0.993	0.995	0.987	0.987
0.226	0.159	0.155	0.166	0.155	0.150	0.155	0.153	0.147	0.153	0.154
0.238	0.434	0.437	0.431	0.434	0.434	0.436	0.411	0.441	0.435	0.439
0.251	0.423	0.399	0.408	0.403	0.406	0.408	0.438	0.404	0.393	0.383
0.269	0.358	0.343	0.346	0.351	0.347	0.354	0.367	0.358	0.346	0.334
0.286	0.267	0.283	0.264	0.253	0.250	0.228	0.230	0.232	0.245	0.244
0.292	-0.406	-0.405	-0.404	-0.406		-0.411	-0.410	-0.481	-0.411	-0.407
0.322	-0.144	0.014	-0.009	-0.201	-0.207	-0.144	-0.218	-0.213	-0.069	0.144
0.352	-0.121	-0.214	-0.185	-0.113	-0.135	-0.146	-0.146	-0.129	-0.186	0.027
0.381	0.032	0.015		0.041	0.072	0.065	0.075	0.077	0.065	0.021
0.392	0.190	0.179	0.020	0.191	0.193	0.188	0.196	0.193	0.184	0.171
0.402	0.025	-0.011	-0.004	0.013	0.015	0.022	0.018	0.018	0.021	0.018
0.447	0.007	0.034	0.034	0.038	0.020		0.020	0.022	0.020	0.020
0.492	0.027	0.028	0.022	0.024	0.025	0.026	0.028	0.024	0.037	0.039
0.529	0.036	0.032	0.031	0.029	0.032	0.030	0.031	0.030	0.043	0.042
0.592	0.041	0.038	0.033	0.030	0.031	0.034	0.035	0.032	0.082	0.034
0.655	0.059	0.022	0.001	0.015	0.007	0.037	0.013	0.006	0.021	0.014
0.661	-0.072	-0.108	-0.112	-0.095	0.004	-0.078		-0.112	-0.085	-0.114
0.687	-0.080	0.008	-0.069	-0.077	-0.059	-0.080	0.003	-0.069		-0.062

(d) $\alpha = 1.9°$

0.223	1.034	1.034	1.023	1.001	0.990	0.937	0.946	0.959	0.970	0.986
0.226	0.228	0.225	0.225	0.178	0.155	0.085	0.086	0.099	0.129	0.152
0.238	0.463	0.464	0.449	0.441	0.432	0.413	0.388	0.423	0.424	0.436
0.251	0.459	0.432	0.432	0.412	0.402	0.375	0.408	0.381	0.383	0.381
0.269	0.389	0.369	0.367	0.355	0.341	0.331	0.340	0.339	0.328	0.327
0.286	0.299	0.312	0.283	0.256	0.244	0.186	0.195	0.207	0.231	0.237
0.292	-0.396	-0.396	-0.399	-0.406		-0.421	-0.419	-0.492	-0.417	-0.329
0.322	-0.116	0.087	0.064	-0.195	-0.129	-0.178	-0.231	-0.136	-0.096	0.221
0.352	-0.106	-0.217	-0.178	-0.110	-0.057	-0.151	-0.154	-0.130	-0.188	-0.201
0.381	0.003	0.015		0.004	0.067	0.095	0.104	0.093	0.046	0.020
0.392	0.200	0.181	0.198	0.192	0.191	0.178	0.184	0.184	0.194	0.165
0.402	0.014	0.001	-0.031	0.013	0.014	0.025	0.021	0.020	-0.019	0.027
0.447	0.044	0.040	0.039	0.038	0.033		0.028	0.024	0.033	0.012
0.492	0.038	0.035	0.031	0.027	0.018	0.021	0.021	0.021	0.023	0.019
0.529	0.050	0.050	0.042	0.029	0.026	0.024	0.026	0.024	0.024	0.025
0.592	0.056	0.052	0.045	0.035	0.027	0.028	0.028	0.030	0.025	0.023
0.655	0.112	0.198	0.072	0.023	-0.000	0.020	0.114	0.028	0.004	0.005
0.661	-0.044	-0.306	-0.066	-0.106	-0.120	-0.109		-0.093	-0.123	-0.118
0.687	-0.085	0.034	-0.099	-0.113	-0.084	-0.073	-0.013	-0.122		-0.073

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TABLE V.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE No. FOR M_∞ = 1.20 - Continued

(e) α = 3.9°

Axial position x/l	Radial position, θ, deg									
	0	30	45	75	90	180	285	315	330	360
0.223	1.085	1.080	1.057	1.015	0.988	0.885	0.898	0.921	0.951	0.983
0.226	0.299	0.292	0.280	0.201	0.156	0.020	0.024	0.047	0.115	0.151
0.238	0.496	0.492	0.466	0.437	0.416	0.389	0.365	0.400	0.401	0.421
0.251	0.492	0.463	0.450	0.413	0.391	0.343	0.365	0.352	0.363	0.370
0.269	0.419	0.395	0.382	0.356	0.344	0.304	0.312	0.314	0.334	0.330
0.286	0.296	0.303	0.278	0.257	0.249	0.206	0.214	0.223	0.242	0.244
0.292	-0.394	-0.392	-0.397	-0.403		-0.417	-0.413	-0.483	-0.412	-0.407
0.322	-0.138	0.043	-0.008	-0.199	-0.205	-0.164	-0.215	-0.210	-0.059	0.115
0.352	-0.122	-0.211	-0.184	-0.117	-0.135	-0.142	-0.138	-0.128	-0.186	-0.312
0.381	0.055	0.022		-0.002	0.041	0.110	0.115	0.097	-0.005	0.014
0.392	0.208	0.179	0.193	0.186	0.182	0.166	0.169	0.173	0.188	0.145
0.402	0.022	0.006	-0.023	-0.000	0.002	0.027	0.023	0.012	-0.030	0.024
0.447	0.054	0.047	0.040	0.033	0.024		0.022	0.018	0.027	0.015
0.492	0.045	0.040	0.032	0.026	0.015	0.020	0.020	0.020	0.021	0.018
0.529	0.051	0.050	0.042	0.030	0.027	0.027	0.028	0.028	0.028	0.028
0.592	0.051	0.049	0.045	0.038	0.032	0.030	0.030	0.033	0.034	0.031
0.655	0.087	0.176	0.063	0.023	0.002	0.040	0.150	0.039	0.016	0.009
0.661	-0.044	-0.307	-0.073	-0.106	-0.123	-0.115		-0.111	-0.156	-0.120
0.687	-0.102	0.019	-0.114	-0.144	-0.117	-0.091	-0.033	-0.141		-0.102

(f) α = 6.8°

0.223	1.132	1.123	1.086	1.021	0.979	0.826	0.841	0.875	0.924	0.970
0.226	0.378	0.365	0.331	0.223	0.153	-0.040	-0.033	-0.000	0.089	0.148
0.238	0.537	0.528	0.487	0.432	0.398	0.371	0.344	0.375	0.374	0.401
0.251	0.531	0.497	0.468	0.410	0.374	0.313	0.346	0.322	0.340	0.353
0.269	0.452	0.424	0.397	0.345	0.308	0.274	0.282	0.284	0.269	0.286
0.286	0.350	0.357	0.300	0.236	0.204	0.115	0.128	0.144	0.177	0.196
0.292	-0.379	-0.383	-0.397	-0.421		-0.445	-0.449	-0.527	-0.448	-0.434
0.322	-0.044	0.179	0.018	-0.187	-0.239	-0.208	-0.261	-0.253	-0.169	0.075
0.352	-0.077	-0.220	-0.153	-0.165	-0.142	-0.155	-0.162	-0.151	-0.175	-0.365
0.381	-0.017	0.012		-0.111	-0.000	0.118	0.118	0.097	-0.025	0.011
0.392	0.211	0.177	0.181	0.171	0.164	0.157	0.156	0.158	0.170	0.125
0.402	0.009	-0.000	-0.011	-0.034	-0.020	0.029	0.024	-0.012	0.004	0.002
0.447	0.066	0.055	0.034	0.018	0.005		0.015	0.006	-0.000	-0.008
0.492	0.064	0.054	0.028	0.001	-0.016	0.013	0.009	0.001	-0.012	-0.014
0.529	0.081	0.073	0.042	0.003	-0.007	0.016	0.013	0.002	-0.012	-0.011
0.592	0.084	0.073	0.040	0.004	-0.012	0.018	0.010	0.002	-0.016	-0.018
0.655	0.161	0.218	0.081	0.007	-0.033	0.035	0.074	-0.018	-0.027	-0.029
0.661	-0.012	-0.303	-0.074	-0.132	-0.150	-0.125		-0.105	-0.159	-0.154
0.687	-0.079	0.046	-0.090	-0.157	-0.144	-0.065	-0.021	-0.116		-0.116

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TABLE V.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE NO. FOR $M_\infty = 1.20$ - Concluded

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	10	45	72	90	180	200	225	270	270
0.223	1.182	1.170	1.117	1.031	0.972	0.770	0.785	0.833	0.900	0.963
0.226	0.470	0.450	0.392	0.252	0.160	-0.096	-0.085	-0.039	0.071	0.152
0.238	0.578	0.565	0.504	0.416	0.368	0.355	0.325	0.337	0.337	0.369
0.251	0.572	0.533	0.487	0.403	0.351	0.288	0.318	0.291	0.312	0.332
0.269	0.491	0.458	0.415	0.340	0.293	0.249	0.256	0.254	0.244	0.268
0.286	0.372	0.377	0.308	0.225	0.184	0.094	0.105	0.118	0.152	0.175
0.292	-0.364	-0.370	-0.392	-0.425		-0.451	-0.455	-0.442	-0.458	-0.443
0.322	-0.000	0.189	0.027	-0.178	-0.229	-0.217	-0.270	-0.266	-0.153	-0.024
0.352	-0.059	-0.213	-0.127	-0.198	-0.131	-0.146	-0.155	-0.161	-0.136	-0.353
0.381	-0.000	0.004		-0.104	-0.070	0.123	0.116	0.097	-0.018	-0.001
0.392	0.218	0.185	0.168	0.139	0.141	0.156	0.143	0.143	0.122	0.097
0.402	0.015	-0.005	-0.016	-0.060	-0.056	0.030	0.024	-0.050	-0.024	-0.037
0.447	0.080	0.065	0.030	0.005	-0.012		0.011	-0.000	-0.012	-0.026
0.492	0.079	0.064	0.028	-0.012	-0.032	0.013	0.005	-0.006	-0.028	-0.031
0.529	0.096	0.084	0.041	-0.012	-0.027	0.014	0.006	-0.004	-0.029	-0.032
0.592	0.102	0.085	0.039	-0.014	-0.035	0.011	-0.001	-0.007	-0.039	-0.044
0.655	0.185	0.223	0.080	-0.008	-0.054	0.066	0.066	-0.032	-0.049	-0.055
0.661	0.005	-0.297	-0.079	-0.149	-0.168	-0.100		-0.113	-0.181	-0.179
0.687	-0.071	0.052	-0.084	-0.176	-0.173	-0.068	-0.015	-0.117		-0.147

(h) $\alpha = 10.0°$

0.223	1.226	1.214	1.146	1.034	0.958	0.707	0.726	0.785	0.870	0.950
0.226	0.566	0.540	0.455	0.283	0.160	-0.154	-0.140	-0.085	0.049	0.154
0.238	0.623	0.595	0.521	0.393	0.324	0.339	0.302	0.301	0.289	0.325
0.251	0.613	0.569	0.499	0.388	0.320	0.261	0.291	0.256	0.276	0.300
0.269	0.529	0.492	0.429	0.339	0.296	0.217	0.228	0.217	0.253	0.274
0.286	0.360	0.360	0.301	0.232	0.197	0.109	0.126	0.143	0.179	0.197
0.292	-0.371	-0.373	-0.318	-0.424		-0.450	-0.452	-0.395	-0.450	-0.374
0.322	-0.070	0.135	0.005	-0.193	-0.172	-0.201	-0.253	-0.245	-0.132	0.039
0.352	-0.098	-0.223	-0.172	-0.133	-0.102	-0.144	-0.163	-0.141	-0.059	-0.192
0.381	0.070	0.008		-0.059	-0.092	0.197	0.102	0.092	0.048	0.035
0.392	0.229	0.192	0.150	0.080	0.100	0.158	0.122	0.115	0.053	0.058
0.402	0.017	-0.012	-0.033	-0.095	-0.098	0.015	0.015	-0.084	-0.041	-0.082
0.447	0.092	0.071	0.022	-0.019	-0.037		-0.000	-0.011	-0.031	-0.048
0.492	0.086	0.068	0.025	-0.024	-0.045	0.009	-0.000	-0.010	-0.033	-0.035
0.529	0.091	0.079	0.040	-0.009	-0.022	0.013	0.005	-0.002	-0.020	-0.020
0.592	0.084	0.072	0.039	0.002	-0.013	0.016	0.009	0.003	-0.013	-0.013
0.655	0.145	0.211	0.077	0.010	-0.025	0.020	0.088	-0.007	-0.016	-0.016
0.661	-0.028	-0.306	-0.072	-0.124	-0.138	-0.121		-0.097	-0.143	-0.136
0.687	-0.089	0.032	-0.101	-0.130	-0.103	-0.078	-0.020	-0.127		-0.088

[redacted]

[redacted]

TABLE VI.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE NO. FOR $M_\infty = 1.30$

(a) $\alpha = -4.2°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	0.930	0.941	0.963	0.999	1.035	1.136	1.135	1.115	1.066	1.037
0.226	0.104	0.099	0.136	0.173	0.210	0.356	0.346	0.307	0.267	0.216
0.238	0.316	0.310	0.311	0.336	0.360	0.448	0.426	0.419	0.375	0.352
0.251	0.311	0.296	0.309	0.325	0.350	0.434	0.442	0.399	0.356	0.321
0.269	0.295	0.276	0.291	0.328	0.342	0.422	0.439	0.406	0.371	0.339
0.286	0.188	0.207	0.206	0.223	0.241	0.291	0.290	0.300	0.261	0.238
0.292	-0.339	-0.337	-0.329	-0.320		-0.291	-0.292	-0.349	-0.307	-0.238
0.322	-0.143	0.009	0.006	-0.170	0.087	-0.111	-0.141	-0.149	-0.014	0.228
0.352	-0.112	-0.181	-0.177	-0.100	-0.014	-0.104	-0.074	-0.087	-0.178	-0.200
0.381	0.034	-0.022		0.001	-0.005	0.013	0.002	0.011	0.029	-0.024
0.392	0.169	0.157	0.164	0.174	0.178	0.170	0.183	0.181	0.167	0.154
0.402	0.024	-0.009	0.004	0.021	0.025	0.035	0.033	0.032	0.031	0.007
0.447	0.030	0.028	0.028	0.031	0.032		0.033	0.029	0.030	0.024
0.492	0.025	0.026	0.027	0.028	0.024	0.030	0.031	0.033	0.033	0.031
0.529	0.037	0.039	0.037	0.035	0.035	0.036	0.039	0.038	0.037	0.036
0.592	0.036	0.036	0.038	0.040	0.039	0.037	0.038	0.037	0.035	0.032
0.655	0.077	0.173	0.063	0.032	0.021	0.077	0.189	0.061	0.032	0.025
0.661	-0.044	-0.277	-0.042	-0.084	-0.085	-0.041		-0.038	-0.087	-0.087
0.687	-0.072	0.025	-0.100	-0.064	-0.044	-0.077	0.025	-0.102		-0.044

(b) $\alpha = -2.2°$

0.223	0.980	0.986	0.997	1.017	1.036	1.086	1.089	1.080	1.052	1.039
0.226	0.158	0.153	0.175	0.188	0.206	0.286	0.278	0.255	0.241	0.214
0.238	0.352	0.349	0.347	0.366	0.373	0.408	0.386	0.394	0.373	0.366
0.251	0.339	0.322	0.334	0.343	0.355	0.392	0.401	0.373	0.349	0.326
0.269	0.323	0.305	0.315	0.333	0.342	0.386	0.399	0.380	0.360	0.335
0.286	0.201	0.218	0.213	0.221	0.233	0.259	0.259	0.251	0.249	0.234
0.292	-0.334	-0.332	-0.327	-0.321		-0.308	-0.307	-0.362	-0.314	-0.315
0.322	-0.141	0.024	0.006	-0.167	-0.163	-0.133	-0.163	-0.164	-0.034	0.159
0.352	-0.109	-0.193	-0.175	-0.096	-0.099	-0.115	-0.097	-0.102	-0.180	-0.254
0.381	-0.010	-0.027		-0.016	-0.026	-0.018	-0.047	-0.033	0.017	-0.019
0.392	0.157	0.149	0.146	0.163	0.167	0.168	0.181	0.180	0.156	0.138
0.402	0.029	-0.037	0.011	0.010	0.016	0.028	0.029	0.025	0.023	0.029
0.447	0.018	0.017	0.015	0.023	0.022		0.032	0.025	0.024	0.014
0.492	0.013	0.016	0.013	0.017	0.014	0.031	0.032	0.030	0.026	0.019
0.529	0.025	0.027	0.024	0.024	0.026	0.036	0.039	0.035	0.030	0.026
0.592	0.025	0.025	0.026	0.029	0.029	0.035	0.035	0.034	0.028	0.022
0.655	0.061	0.162	0.048	0.021	0.013	0.076	0.187	0.056	0.025	0.016
0.661	-0.055	-0.258	-0.053	-0.093	-0.094	-0.045		-0.045	-0.096	-0.093
0.687	-0.079	0.020	-0.106	-0.072	-0.052	-0.084	0.015	-0.110		-0.053

[redacted]

[redacted]

TABLE VI.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_3 FOR $M_\infty = 1.30$ - Continued

(c) $\alpha = -0.1°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	75	90	180	200	225	252	270
0.223	1.036	1.038	1.037	1.035	1.037	1.037	1.041	1.046	1.040	1.039
0.226	0.223	0.215	0.223	0.208	0.210	0.211	0.209	0.206	0.219	0.215
0.238	0.384	0.383	0.374	0.379	0.380	0.371	0.354	0.372	0.368	0.373
0.251	0.375	0.357	0.363	0.359	0.358	0.354	0.371	0.351	0.340	0.328
0.269	0.362	0.340	0.344	0.344	0.345	0.352	0.362	0.356	0.354	0.337
0.286	0.229	0.242	0.232	0.232	0.236	0.238	0.241	0.239	0.246	0.236
0.292	-0.322	-0.320	-0.319	-0.317		-0.318	-0.315	-0.371	-0.315	-0.313
0.322	-0.135	-0.138	0.009	-0.164	-0.162	-0.137	-0.166	-0.166	-0.034	0.161
0.352	-0.107	-0.193	-0.172	-0.093	-0.098	-0.115	-0.096	-0.101	-0.178	-0.246
0.381	-0.005	-0.029		-0.011	-0.008	0.012	-0.007	0.001	0.025	-0.032
0.392	0.163	0.151	0.159	0.169	0.173	0.163	0.176	0.174	0.161	0.148
0.402	0.018	-0.015	-0.000	0.018	0.021	0.030	0.032	0.027	0.027	0.015
0.447	0.025	0.023	0.024	0.027	0.027		0.027	0.023	0.024	0.017
0.492	0.020	0.021	0.021	0.022	0.019	0.024	0.026	0.027	0.027	0.025
0.529	0.031	0.033	0.032	0.029	0.030	0.031	0.034	0.033	0.032	0.031
0.592	0.033	0.031	0.033	0.034	0.033	0.032	0.032	0.032	0.030	0.026
0.655	0.071	0.166	0.056	0.025	0.015	0.072	0.183	0.055	0.025	0.019
0.661	-0.051	-0.248	-0.050	-0.091	-0.093	-0.046		-0.045	-0.094	-0.094
0.687	-0.079	0.020	-0.106	-0.069	-0.050	-0.084	0.018	-0.109		-0.051

(d) $\alpha = 1.9°$

0.223	1.082	1.081	1.069	1.049	1.036	0.986	0.997	1.010	1.019	1.037
0.226	0.290	0.280	0.272	0.229	0.209	0.149	0.150	0.160	0.196	0.211
0.238	0.417	0.413	0.395	0.384	0.375	0.344	0.320	0.346	0.357	0.371
0.251	0.409	0.386	0.383	0.364	0.352	0.322	0.339	0.329	0.331	0.329
0.269	0.395	0.371	0.368	0.354	0.338	0.321	0.329	0.331	0.324	0.326
0.286	0.286	0.300	0.270	0.242	0.228	0.181	0.185	0.193	0.221	0.228
0.292	-0.300	-0.301	-0.304	-0.313		-0.340	-0.338	-0.390	-0.326	-0.317
0.322	-0.105	0.100	0.016	-0.161	-0.171	-0.157	-0.193	-0.191	-0.080	0.153
0.352	-0.094	-0.201	-0.165	-0.101	-0.100	-0.126	-0.124	-0.114	-0.193	-0.263
0.381	-0.033	-0.017		-0.052	-0.019	0.039	0.043	0.031	-0.004	-0.027
0.392	0.168	0.153	0.170	0.165	0.168	0.161	0.169	0.167	0.171	0.140
0.402	0.017	-0.001	-0.024	0.017	0.018	0.030	0.027	0.026	-0.004	0.028
0.447	0.033	0.029	0.026	0.023	0.023		0.028	0.017	0.023	0.013
0.492	0.029	0.027	0.025	0.011	0.013	0.015	0.017	0.017	0.029	0.015
0.529	0.042	0.042	0.036	0.024	0.016	0.010	0.022	0.010	0.030	0.010
0.592	0.045	0.042	0.036	0.028	0.021	0.028	0.018	0.027	0.017	0.015
0.655	0.113	0.197	0.069	0.029	0.014	0.035	0.125	0.037	0.022	0.009
0.661	-0.029	-0.244	-0.046	-0.098	-0.102	-0.074		-0.049	-0.108	-0.090
0.687	-0.094	0.039	-0.098	-0.088	-0.066	-0.074	-0.010	-0.117		-0.242

[redacted]

TABLE VI.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_5 FOR $M_\infty = 1.30$ - Continued

(e) $\alpha = 3.9°$

Axial position x/l	Radial position, θ, deg									
	0	9	45	72	90	180	200	225	252	270
0.223	1.129	1.125	1.100	1.060	1.031	0.931	0.943	0.968	0.999	1.031
0.226	0.352	0.342	0.317	0.248	0.205	0.086	0.088	0.107	0.171	0.205
0.238	0.449	0.443	0.412	0.377	0.352	0.304	0.278	0.311	0.324	0.347
0.251	0.446	0.417	0.402	0.365	0.339	0.288	0.307	0.301	0.313	0.318
0.269	0.429	0.402	0.386	0.355	0.331	0.290	0.296	0.304	0.313	0.322
0.286	0.324	0.336	0.299	0.255	0.231	0.159	0.188	0.197	0.207	0.223
0.292	-0.287	-0.288	-0.296	-0.312		-0.353	-0.352	-0.404	-0.337	-0.324
0.322	-0.083	0.118	0.020	-0.163	-0.182	-0.161	-0.181	-0.205	-0.111	0.131
0.352	-0.084	-0.202	-0.154	-0.117	-0.105	-0.131	-0.134	-0.119	-0.198	-0.291
0.381	-0.026	-0.012		-0.076	-0.051	0.065	0.071	0.042	-0.038	-0.011
0.392	0.170	0.153	0.168	0.150	0.152	0.151	0.154	0.154	0.163	0.119
0.402	0.014	0.003	-0.026	0.019	0.005	0.027	0.024	0.017	0.051	0.021
0.447	0.041	0.032	0.024	0.017	0.012		0.015	0.010	0.017	0.004
0.492	0.038	0.035	0.026	0.017	0.006	0.011	0.012	0.012	0.009	0.008
0.529	0.051	0.049	0.036	0.019	0.013	0.018	0.018	0.015	0.011	0.012
0.592	0.054	0.050	0.037	0.021	0.012	0.015	0.012	0.011	0.005	0.004
0.655	0.133	0.208	0.075	0.016	-0.004	0.032	0.108	0.016	-0.000	-0.000
0.661	0.001	-0.241	-0.043	-0.102	-0.112	-0.084		-0.067	-0.119	-0.109
0.687	-0.098	0.045	-0.097	-0.106	-0.084	-0.071	-0.012	-0.112		-0.081

(f) $\alpha = 6.0°$

0.223	1.177	1.169	1.132	1.071	1.026	0.880	0.893	0.929	0.976	1.022
0.226	0.417	0.406	0.365	0.272	0.213	0.037	0.036	0.065	0.149	0.204
0.238	0.492	0.484	0.437	0.375	0.328	0.277	0.242	0.268	0.289	0.327
0.251	0.490	0.455	0.423	0.365	0.325	0.264	0.281	0.276	0.295	0.307
0.269	0.468	0.435	0.406	0.356	0.323	0.264	0.268	0.278	0.296	0.312
0.286	0.358	0.364	0.318	0.260	0.229	0.145	0.148	0.158	0.200	0.222
0.292	-0.273	-0.275	-0.288	-0.309		-0.359	-0.358	-0.411	-0.339	-0.324
0.322	-0.071	0.150	0.022	-0.161	-0.183	-0.161	-0.202	-0.208	-0.116	0.080
0.352	-0.080	-0.200	-0.150	-0.121	-0.107	-0.130	-0.135	-0.120	-0.202	-0.343
0.381	-0.023	-0.011		-0.078	-0.090	0.084	0.086	0.048	-0.046	-0.024
0.392	0.182	0.163	0.171	0.131	0.131	0.147	0.144	0.144	0.152	0.103
0.402	0.022	0.009	-0.013	-0.013	-0.009	0.029	0.026	0.001	-0.005	0.009
0.447	0.056	0.044	0.023	0.005	-0.001		0.013	0.005	-0.002	-0.008
0.492	0.055	0.048	0.028	0.009	-0.004	0.011	0.009	0.008	-0.000	-0.002
0.529	0.064	0.059	0.038	0.010	0.002	0.017	0.016	0.010	0.001	0.002
0.592	0.065	0.058	0.038	0.017	0.005	0.015	0.011	0.007	-0.034	-0.000
0.655	0.144	0.214	0.079	0.014	-0.008	0.035	0.106	0.011	-0.000	-0.001
0.661	0.009	-0.236	-0.041	-0.103	-0.114	-0.083		-0.063	-0.120	-0.110
0.687	-0.096	0.048	-0.095	-0.107	-0.085	-0.068	-0.010	-0.110		-0.082

[redacted]

TABLE VI.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE NO. 508 FOR M_∞=1.30 - Concluded

(g) α = 8.0°

Axial position x/l	Radial position, θ, deg									
	0	30	45	75	90	180	270	285	315	270
0.223	1.220	1.210	1.159	1.079	1.018	0.824	0.838	0.883	0.948	1.012
0.226	0.481	0.467	0.409	0.291	0.214	-0.015	-0.007	0.031	0.123	0.197
0.238	0.538	0.524	0.460	0.367	0.305	0.251	0.199	0.210	0.254	0.301
0.251	0.533	0.493	0.442	0.356	0.301	0.236	0.252	0.244	0.267	0.284
0.269	0.502	0.467	0.423	0.348	0.303	0.235	0.240	0.246	0.262	0.288
0.286	0.396	0.397	0.337	0.258	0.216	0.116	0.117	0.129	0.179	0.208
0.292	-0.259	-0.262	-0.283	-0.314		-0.370	-0.370	-0.426	-0.351	-0.334
0.322	-0.042	0.193	0.030	-0.165	-0.201	-0.167	-0.213	-0.227	-0.154	0.032
0.352	-0.062	-0.196	-0.132	-0.147	-0.133	-0.134	-0.142	-0.134	-0.208	-0.322
0.381	-0.008	-0.009		-0.095	-0.132	0.093	0.088	0.050	-0.058	-0.044
0.392	0.201	0.172	0.167	0.102	0.093	0.145	0.131	0.128	0.095	0.073
0.402	0.022	0.012	-0.004	-0.052	-0.080	0.029	-0.151	-0.028	-0.017	-0.068
0.447	0.070	0.055	0.020	-0.013	-0.024		0.005	-0.003	-0.018	-0.032
0.492	0.070	0.059	0.027	-0.009	-0.026	0.009	0.002	-0.006	-0.022	-0.025
0.529	0.081	0.071	0.035	-0.007	-0.019	0.014	0.009	-0.001	-0.020	-0.021
0.592	0.081	0.071	0.036	-0.000	-0.017	0.012	0.004	-0.001	-0.020	-0.024
0.655	0.168	0.220	0.080	-0.000	-0.028	0.059	0.095	-0.004	-0.020	-0.024
0.661	0.027	-0.232	-0.044	-0.119	-0.134	-0.067		-0.078	-0.138	-0.130
0.687	-0.091	0.052	-0.267	-0.132	-0.114	-0.069	-0.011	-0.105		-0.106

(h) α = 10.0°

0.223	1.264	1.249	1.187	1.084	1.008	0.767	0.785	0.838	0.919	0.997
0.226	0.559	0.538	0.462	0.315	0.217	-0.067	-0.055	-0.009	0.100	0.194
0.238	0.588	0.563	0.487	0.361	0.280	0.220	0.160	0.152	0.213	0.278
0.251	0.580	0.536	0.462	0.343	0.263	0.210	0.229	0.212	0.231	0.247
0.269	0.539	0.497	0.438	0.344	0.294	0.208	0.217	0.210	0.247	0.276
0.286	0.410	0.406	0.342	0.257	0.211	0.111	0.112	0.126	0.176	0.205
0.292	-0.252	-0.255	-0.281	-0.316		-0.373	-0.370	-0.428	-0.354	-0.337
0.322	-0.044	0.241	0.032	-0.161	-0.195	-0.164	-0.208	-0.220	-0.134	-0.018
0.352	-0.067	-0.194	-0.138	-0.128	-0.166	-0.127	-0.133	-0.121	-0.191	-0.234
0.381	0.015	-0.009		-0.070	-0.165	0.097	0.080	0.053	-0.088	-0.100
0.392	0.228	0.186	0.147	0.076	0.026	0.152	0.112	0.103	0.039	0.041
0.402	0.038	0.016	-0.004	-0.098	-0.084	0.022	-0.163	-0.071	-0.039	-0.068
0.447	0.088	0.064	0.012	-0.040	-0.050		-0.004	-0.014	-0.038	-0.058
0.492	0.085	0.069	0.027	-0.026	-0.045	0.010	-0.003	-0.012	-0.036	-0.041
0.529	0.092	0.079	0.037	-0.018	-0.032	0.015	0.003	-0.004	-0.029	-0.031
0.592	0.087	0.077	0.038	-0.002	-0.021	0.012	0.004	-0.001	-0.020	-0.024
0.655	0.166	0.220	0.081	0.002	-0.028	0.054	0.099	0.001	-0.014	-0.016
0.661	0.048	-0.231	-0.041	-0.112	-0.158	-0.048		-0.067	-0.127	-0.152
0.687	-0.091	0.051	-0.267	-0.113	-0.091	-0.068	-0.006	-0.106		-0.082

[redacted]

[redacted]

TABLE VII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N[illegible] FOR M_∞=1.45

(a) α = -4.2°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	0.989	1.001	1.024	1.060	1.097	1.197	1.196	1.176	1.127	1.099
0.226	0.173	0.170	0.205	0.240	0.287	0.413	0.404	0.365	0.322	0.280
0.238	0.205	0.214	0.230	0.267	0.299	0.408	0.388	0.385	0.336	0.306
0.251	0.265	0.257	0.263	0.279	0.306	0.399	0.415	0.365	0.315	0.280
0.269	0.263	0.246	0.260	0.282	0.302	0.395	0.414	0.376	0.339	0.295
0.286	0.166	0.181	0.182	0.200	0.223	0.309	0.304	0.275	0.245	0.215
0.292	-0.265	-0.261	-0.256	-0.245		-0.196	-0.198	-0.212	-0.227	-0.237
0.322	-0.136	-0.003	0.008	-0.144	-0.132	-0.085	-0.095	-0.106	-0.000	0.163
0.352	-0.097	-0.161	-0.165	-0.074	-0.064	-0.081	-0.045	-0.044	-0.151	-0.230
0.381	0.002	-0.027		-0.070	-0.086	-0.011	-0.048	-0.090	-0.045	-0.046
0.392	0.128	0.123	0.114	0.124	0.112	0.156	0.140	0.132	0.116	0.091
0.402	0.020	-0.038	0.011	-0.165	0.010	0.028	0.027	0.023	0.003	0.025
0.447	0.022	0.022	0.007	0.014	0.024		0.037	0.027	0.020	0.011
0.492	0.014	0.014	0.009	0.009	0.007	0.042	0.040	0.032	0.018	0.009
0.529	0.016	0.017	0.014	0.012	0.016	0.051	0.052	0.039	0.024	0.015
0.592	0.017	0.017	0.015	0.025	0.016	0.053	0.050	0.040	0.022	0.011
0.655	0.048	0.126	0.029	0.012	0.009	0.103	0.210	0.061	0.025	0.010
0.661	-0.038	-0.224	-0.050	-0.085	-0.081	0.008		-0.022	-0.078	-0.079
0.687	-0.077	0.010	-0.097	-0.058	-0.054	-0.094	0.043	-0.092		-0.052

(b) α = -2.2°

0.223	1.039	1.045	1.057	1.077	1.097	1.145	1.147	1.138	1.108	1.097
0.226	0.226	0.218	0.242	0.253	0.285	0.338	0.331	0.312	0.296	0.273
0.238	0.247	0.252	0.254	0.284	0.295	0.353	0.335	0.344	0.316	0.303
0.251	0.292	0.279	0.284	0.294	0.308	0.353	0.370	0.333	0.305	0.281
0.269	0.287	0.270	0.281	0.299	0.307	0.353	0.365	0.344	0.323	0.297
0.286	0.202	0.220	0.213	0.215	0.224	0.245	0.244	0.233	0.229	0.216
0.292	-0.249	-0.248	-0.243	-0.240		-0.227	-0.227	-0.271	-0.237	-0.237
0.322	-0.124	0.038	0.013	-0.134	-0.130	-0.119	-0.128	-0.129	-0.025	0.163
0.352	-0.094	-0.174	-0.154	-0.075	-0.057	-0.101	-0.068	-0.076	-0.148	-0.216
0.381	-0.026	-0.025		-0.058	-0.081	-0.024	-0.059	-0.072	-0.026	-0.047
0.392	0.128	0.123	0.118	0.130	0.128	0.139	0.137	0.134	0.119	0.108
0.402	0.022	-0.030	0.010	0.011	0.016	0.020	0.021	0.019	0.012	0.026
0.447	0.018	0.018	0.005	0.018	0.019		0.023	0.020	0.023	0.016
0.492	0.013	0.024	0.012	0.022	0.012	0.022	0.023	0.022	0.019	0.008
0.529	0.016	0.019	0.017	0.018	0.021	0.030	0.033	0.029	0.024	0.020
0.592	0.019	0.019	0.021	0.023	0.022	0.028	0.028	0.027	0.022	0.016
0.655	0.060	0.155	0.040	0.018	0.011	0.069	0.175	0.046	0.022	0.009
0.661	-0.033	-0.201	-0.043	-0.081	-0.078	-0.025		-0.035	-0.082	-0.076
0.687	-0.092	0.019	-0.099	-0.055	-0.048	-0.086	0.035	-0.100		-0.050

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TABLE VII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_4 FOR M_∞=1.49 - Continued

(c) α = -0.1°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	1.095	1.099	1.100	1.099	1.104	1.100	1.106	1.107	1.100	1.101
0.226	0.289	0.283	0.290	0.278	0.290	0.278	0.276	0.272	0.282	0.281
0.238	0.304	0.305	0.299	0.306	0.303	0.306	0.294	0.314	0.307	0.308
0.251	0.332	0.315	0.314	0.314	0.315	0.319	0.340	0.316	0.304	0.292
0.269	0.326	0.305	0.309	0.312	0.314	0.323	0.331	0.324	0.323	0.304
0.286	0.226	0.239	0.229	0.226	0.230	0.240	0.241	0.233	0.234	0.224
0.292	-0.236	-0.234	-0.233	-0.232		-0.229	-0.227	-0.233	-0.232	-0.232
0.322	-0.120	0.034	0.020	-0.128	-0.124	-0.111	-0.119	-0.120	-0.009	0.172
0.352	-0.091	-0.166	-0.151	-0.067	-0.052	-0.091	-0.053	-0.057	-0.143	-0.207
0.381	-0.008	-0.025		-0.060	-0.074	-0.018	-0.053	-0.051	-0.015	-0.038
0.392	0.137	0.132	0.132	0.135	0.139	0.137	0.142	0.136	0.129	0.121
0.402	0.023	-0.008	0.008	0.019	0.024	0.025	0.024	0.021	0.022	0.016
0.447	0.027	0.027	0.027	0.028	0.027		0.023	0.023	0.029	0.026
0.492	0.023	0.023	0.023	0.024	0.020	0.021	0.022	0.025	0.024	0.022
0.529	0.027	0.029	0.028	0.026	0.027	0.031	0.033	0.032	0.030	0.027
0.592	0.028	0.027	0.030	0.031	0.030	0.032	0.032	0.031	0.028	0.024
0.655	0.067	0.161	0.047	0.025	0.017	0.079	0.188	0.054	0.029	0.020
0.661	-0.031	-0.194	-0.038	-0.075	-0.072	-0.011		-0.026	-0.074	-0.071
0.687	-0.081	0.019	-0.095	-0.050	-0.044	-0.088	0.035	-0.092		-0.043

(d) α = 1.9°

-0.223	1.142	1.141	1.130	1.108	1.097	1.048	1.056	1.071	1.080	1.096
0.226	0.345	0.333	0.328	0.290	0.285	0.218	0.219	0.224	0.260	0.276
0.238	0.349	0.348	0.328	0.316	0.298	0.256	0.245	0.277	0.289	0.304
0.251	0.364	0.344	0.334	0.317	0.305	0.283	0.307	0.289	0.289	0.287
0.269	0.359	0.336	0.334	0.323	0.309	0.287	0.294	0.294	0.297	0.296
0.286	0.273	0.287	0.261	0.235	0.224	0.186	0.188	0.191	0.211	0.216
0.292	-0.213	-0.215	-0.219	-0.228		-0.256	-0.254	-0.295	-0.245	-0.236
0.322	-0.091	0.100	0.021	-0.126	-0.133	-0.137	-0.149	-0.147	-0.056	0.159
0.352	-0.074	-0.173	-0.140	-0.087	-0.066	-0.105	-0.084	-0.091	-0.161	-0.225
0.381	-0.020	-0.017		-0.058	-0.078	-0.006	-0.034	-0.040	-0.029	-0.048
0.392	0.141	0.134	0.147	0.126	0.132	0.130	0.136	0.129	0.136	0.113
0.402	0.013	0.004	-0.006	0.016	0.022	0.023	0.020	0.020	0.001	0.015
0.447	0.033	0.030	0.028	0.026	0.023		0.018	0.017	0.027	0.009
0.492	0.031	0.029	0.025	0.022	0.015	0.014	0.014	0.016	0.023	0.015
0.529	0.036	0.037	0.032	0.022	0.008	0.020	0.022	0.020	0.015	0.018
0.592	0.039	0.037	0.033	0.027	0.020	0.010	0.018	0.017	0.016	0.014
0.655	0.096	0.184	0.056	0.020	0.007	0.044	0.132	0.037	0.013	0.011
0.661	-0.000	-0.185	-0.033	-0.081	-0.083	-0.044		-0.054	-0.088	-0.075
0.687	-0.098	0.040	-0.093	-0.063	-0.053	-0.071	-0.004	-0.101		-0.060

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TABLE VII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_5 FOR M_∞ =1.4[illegible] - Continued

(e) α = 3.9°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	288	252	270
0.223	1.188	1.185	1.160	1.121	1.094	0.992	1.004	1.029	1.056	1.090
0.226	0.410	0.398	0.372	0.307	0.283	0.158	0.161	0.179	0.234	0.269
0.238	0.397	0.394	0.362	0.315	0.298	0.206	0.205	0.241	0.269	0.303
0.251	0.404	0.378	0.358	0.322	0.297	0.250	0.275	0.260	0.270	0.276
0.269	0.401	0.374	0.358	0.325	0.302	0.254	0.260	0.265	0.282	0.288
0.286	0.294	0.303	0.270	0.234	0.217	0.168	0.171	0.176	0.201	0.210
0.292	-0.205	-0.206	-0.215	-0.230		-0.268	-0.265	-0.306	-0.253	-0.241
0.322	-0.087	0.105	0.017	-0.129	-0.138	-0.143	-0.157	-0.154	-0.064	0.130
0.352	-0.074	-0.175	-0.140	-0.093	-0.073	-0.108	-0.089	-0.097	-0.169	-0.261
0.381	-0.020	-0.018		-0.062	-0.088	0.013	-0.171	-0.033	-0.041	-0.058
0.392	0.148	0.136	0.152	0.113	0.111	0.123	0.126	0.119	0.131	0.092
0.402	0.012	0.012	-0.009	0.003	0.005	0.020	0.018	0.014	-0.015	0.014
0.447	0.037	0.031	0.025	0.017	0.011		0.012	0.007	0.028	0.007
0.492	0.035	0.032	0.022	0.014	0.004	0.008	0.008	0.008	0.004	0.005
0.529	0.043	0.041	0.031	0.016	0.011	0.016	0.016	0.012	0.008	0.009
0.592	0.043	0.039	0.030	0.020	0.013	0.013	0.012	0.011	0.009	0.007
0.655	0.098	0.184	0.054	0.015	-0.000	0.037	0.122	0.031	0.008	0.005
0.661	0.002	-0.187	-0.035	-0.085	-0.088	-0.050		-0.062	-0.093	-0.084
0.687	-0.101	0.039	-0.095	-0.068	-0.057	-0.072	-0.008	-0.104		-0.065

(f) α = 6.0°

0.223	1.237	1.230	1.192	1.131	1.087	0.940	0.952	0.989	1.037	1.083
0.226	0.478	0.464	0.423	0.327	0.282	0.105	0.113	0.142	0.215	0.268
0.238	0.450	0.441	0.396	0.342	0.294	0.163	0.167	0.200	0.251	0.300
0.251	0.450	0.418	0.384	0.323	0.281	0.223	0.248	0.231	0.249	0.265
0.269	0.445	0.414	0.384	0.331	0.290	0.226	0.232	0.238	0.249	0.271
0.286	0.356	0.364	0.308	0.239	0.203	0.131	0.132	0.138	0.171	0.194
0.292	-0.174	-0.180	-0.197	-0.227		-0.282	-0.282	-0.326	-0.265	-0.247
0.322	-0.042	0.148	0.028	-0.133	-0.156	-0.147	-0.171	-0.176	-0.115	0.104
0.352	-0.036	-0.167	-0.114	-0.119	-0.092	-0.106	-0.105	-0.117	-0.210	-0.294
0.381	0.007	-0.005		-0.080	-0.099	0.032	0.027	-0.198	-0.054	-0.067
0.392	0.168	0.145	0.156	0.104	0.082	0.121	0.119	0.114	0.122	0.073
0.402	0.022	0.021	-0.000	-0.018	-0.011	0.026	0.021	0.004	-0.050	-0.005
0.447	0.053	0.041	0.021	0.004	-0.002		0.009	-0.000	-0.003	-0.010
0.492	0.052	0.044	0.015	0.001	-0.012	0.007	0.003	-0.000	-0.011	-0.012
0.529	0.066	0.059	0.035	0.001	-0.008	0.013	0.010	-0.000	-0.012	-0.013
0.592	0.072	0.061	0.033	0.003	-0.012	0.009	0.003	-0.002	-0.016	-0.018
0.655	0.137	0.204	0.060	-0.002	-0.027	0.051	0.103	0.005	-0.018	-0.023
0.661	0.040	-0.176	-0.031	-0.102	-0.113	-0.041		-0.058	-0.117	-0.111
0.687	-0.082	0.055	-0.088	-0.101	-0.089	-0.081	-0.007	-0.097		-0.098

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TABLE VII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE NO. FOR $M_\infty = 1.45$ - Concluded

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.223	1.282	1.270	1.221	1.137	1.080	0.886	0.901	0.947	1.012	1.075
0.226	0.541	0.524	0.470	0.347	0.280	0.056	0.066	0.103	0.195	0.264
0.238	0.503	0.491	0.421	0.345	0.279	0.123	0.124	0.182	0.230	0.286
0.251	0.500	0.463	0.411	0.324	0.263	0.199	0.220	0.200	0.222	0.247
0.269	0.496	0.459	0.411	0.329	0.276	0.201	0.206	0.211	0.223	0.255
0.286	0.400	0.402	0.330	0.236	0.186	0.111	0.110	0.112	0.148	0.179
0.292	-0.154	-0.162	-0.187	-0.229		-0.292	-0.292	-0.341	-0.277	-0.254
0.322	-0.012	0.225	0.035	-0.137	-0.171	-0.147	-0.178	-0.191	-0.147	0.063
0.352	-0.009	-0.160	-0.095	-0.138	-0.129	-0.107	-0.115	-0.134	-0.222	-0.259
0.381	0.025	0.004		-0.111	-0.115	0.044	0.041	-0.020	-0.082	-0.086
0.392	0.190	0.154	0.144	0.094	0.042	0.121	0.109	0.106	0.058	0.038
0.402	0.039	0.024	0.012	-0.043	-0.046	0.020	0.023	-0.023	-0.020	-0.041
0.447	0.068	0.049	0.011	-0.014	-0.026		0.002	-0.008	-0.021	-0.034
0.492	0.066	0.053	0.021	-0.018	-0.034	0.006	-0.003	-0.013	-0.031	-0.034
0.529	0.081	0.070	0.032	-0.015	-0.029	0.011	0.002	-0.008	-0.032	-0.034
0.592	0.087	0.073	0.032	-0.013	-0.033	0.007	-0.007	-0.010	-0.035	-0.040
0.655	0.159	0.213	0.061	-0.019	-0.050	0.060	0.090	-0.008	-0.038	-0.047
0.661	0.061	-0.171	-0.031	-0.118	-0.134	-0.024		-0.063	-0.135	-0.132
0.687	-0.062	0.065	-0.080	-0.123	-0.115	0.026	0.005	-0.092		-0.121

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TABLE VIII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR M_∞=0.70

(a) α = -4.2°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	0.583	0.587	0.622	0.671	0.731	0.877	0.868	0.838	0.776	0.742
0.234	0.300	0.291	0.325	0.373	0.414	0.564	0.563	0.522	0.467	0.423
0.246	0.014	0.013	0.013	0.046	0.075	0.213	0.205	0.173	0.119	0.092
0.251	-0.140	-0.140	-0.116	-0.085	-0.042	0.048	0.040	0.034	-0.013	-0.045
0.268	-0.051	-0.026	-0.038	0.001	0.010	0.091	0.096	0.082	0.025	0.005
0.286	-0.335	-0.332	-0.328	-0.309	-0.294	-0.231	-0.235	-0.225	-0.273	-0.297
0.292	-0.691	-0.690	-0.687	-0.684	-0.668	-0.608	-0.620	-0.728	-0.650	-0.669
0.322	-0.039	-0.030	-0.025	-0.045	-0.043	-0.036	-0.036	-0.037	-0.025	0.059
0.352	0.022	0.083	0.043	0.013	0.013	0.020	0.015	0.011	0.012	0.043
0.381	0.132	0.131		0.127	0.121	0.161	0.157	0.148	0.133	0.102
0.392	0.066	0.052	0.056	0.059	0.063	0.107	0.106	0.090	0.072	0.045
0.402	-0.023	-0.035	-0.030	-0.033	-0.027	0.010	0.002	-0.007	-0.018	-0.018
0.447	0.030	0.024	0.007	0.009	0.011		0.039	0.024	0.016	0.002
0.492	0.011	0.011	0.006	0.007	0.004	0.043	0.041	0.032	0.018	0.007
0.529	0.002	0.006	0.001	-0.002	-0.000	0.033	0.033	0.020	0.007	-0.003
0.592	-0.005	-0.005	-0.007	-0.005	-0.004	0.023	0.022	0.016	0.004	-0.006
0.655	-0.118	-0.108	-0.113	-0.113	-0.114	-0.115	-0.074	-0.117	-0.112	-0.120
0.661	-0.137	-0.204	-0.128	-0.128	-0.124	-0.149		-0.137	-0.130	-0.117
0.687	-0.018	0.034	-0.027	-0.038	-0.030	-0.013	0.049	-0.024		-0.028

(b) α = -2.2°

0.231	0.667	0.667	0.681	0.703	0.740	0.815	0.807	0.797	0.764	0.752
0.234	0.370	0.359	0.379	0.402	0.424	0.497	0.499	0.478	0.456	0.431
0.246	0.064	0.062	0.058	0.065	0.082	0.153	0.149	0.139	0.111	0.096
0.251	-0.085	-0.092	-0.078	-0.062	-0.042	-0.001	-0.002	-0.002	-0.021	-0.032
0.268	-0.022	0.003	-0.011	0.026	0.024	0.059	0.059	0.062	0.021	0.016
0.286	-0.294	-0.291	-0.293	-0.286	-0.283	-0.275	-0.275	-0.251	-0.277	-0.286
0.292	-0.666	-0.668	-0.662	-0.664	-0.658	-0.656	-0.662	-0.761	-0.661	-0.657
0.322	-0.027	0.004	-0.040	-0.039	-0.034	-0.040	-0.038	-0.038	-0.052	0.063
0.352	0.026	0.063	0.054	0.019	0.016	0.021	0.016	0.016	0.032	0.066
0.381	0.148	0.140		0.137	0.132	0.151	0.150	0.149	0.142	0.118
0.392	0.076	0.059	0.068	0.074	0.076	0.096	0.097	0.089	0.082	0.057
0.402	-0.018	-0.028	-0.020	-0.018	-0.013	0.002	-0.001	-0.004	-0.007	-0.005
0.447	0.018	0.019	0.015	0.020	0.022		0.030	0.024	0.024	-0.003
0.492	0.017	0.019	0.016	0.021	0.017	0.029	0.030	0.030	0.027	0.020
0.529	0.012	0.029	0.010	0.009	0.011	0.022	0.024	0.019	-0.002	0.007
0.592	-0.000	-0.000	-0.000	0.003	0.002	0.031	0.016	-0.002	0.007	-0.000
0.655	-0.119	-0.109	-0.112	-0.109	-0.110	-0.116	-0.071	-0.115	-0.109	-0.116
0.661	-0.136	-0.205	-0.127	-0.125	-0.121	-0.148		-0.136	-0.128	-0.115
0.687	-0.017	0.035	-0.025	-0.036	-0.028	-0.012	0.049	-0.023		-0.027

[redacted]

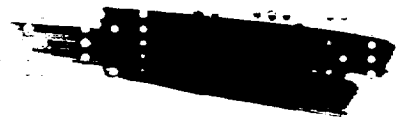

TABLE VIII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 0.70$ - Continued

(c) $\alpha = -0.1°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	0.741	0.736	0.735	0.727	0.745	0.747	0.740	0.745	0.743	0.754
0.234	0.438	0.424	0.430	0.429	0.433	0.433	0.440	0.436	0.439	0.436
0.246	0.120	0.121	0.100	0.089	0.092	0.098	0.098	0.103	0.100	0.105
0.251	-0.032	-0.043	-0.046	-0.044	-0.052	-0.050	-0.046	-0.044	-0.047	-0.034
0.268	0.012	0.026	0.028	0.029	0.026	0.030	0.027	0.040	0.024	0.017
0.286	-0.294	-0.296	-0.296	-0.287	-0.281	-0.260	-0.259	-0.237	-0.272	-0.287
0.292	-0.675	-0.670	-0.672	-0.671	-0.657	-0.624	-0.627	-0.729	-0.645	-0.659
0.322	-0.036	-0.029	-0.018	-0.043	-0.033	-0.029	-0.030	-0.032	-0.014	0.071
0.352	0.026	0.096	0.043	0.011	0.008	0.026	0.020	0.010	0.010	0.014
0.381	0.127	0.127		0.118	0.137	0.144	0.144	0.146	0.146	0.130
0.392	0.086	0.067	0.080	0.082	0.081	0.087	0.089	0.086	0.084	0.065
0.402	-0.013	-0.020	-0.018	-0.012	-0.008	-0.003	-0.006	-0.005	-0.004	0.001
0.447	0.028	0.026	0.025	0.029	0.028		0.028	0.025	0.028	0.020
0.492	0.024	0.024	0.022	0.026	0.021	0.029	0.030	0.032	0.030	0.024
0.529	0.020	0.017	0.017	0.011	0.015	0.015	0.017	0.020	0.015	0.012
0.592	0.008	0.006	0.007	0.008	0.006	0.006	0.008	0.010	0.008	0.003
0.655	-0.119	-0.111	-0.113	-0.105	-0.105	-0.119	-0.076	-0.111	-0.104	-0.113
0.661	-0.140	-0.207	-0.130	-0.122	-0.116	-0.145		-0.130	-0.122	-0.112
0.687	-0.014	0.038	-0.020	-0.032	-0.024	-0.015	0.041	-0.024		-0.023

(d) $\alpha = 1.9°$

0.231	0.806	0.798	0.777	0.740	0.740	0.677	0.671	0.694	0.720	0.751
0.234	0.499	0.480	0.472	0.441	0.423	0.368	0.371	0.386	0.413	0.432
0.246	0.172	0.165	0.134	0.101	0.083	0.044	0.046	0.062	0.077	0.101
0.251	0.010	-0.000	-0.005	-0.032	-0.052	-0.098	-0.088	-0.063	-0.063	-0.036
0.268	0.030	0.052	0.028	0.040	0.025	0.010	0.006	0.015	0.007	0.013
0.286	-0.253	-0.256	-0.268	-0.276	-0.284	-0.309	-0.309	-0.277	-0.291	-0.290
0.292	-0.628	-0.630	-0.637	-0.657	-0.662	-0.681	-0.684	-0.675	-0.676	-0.658
0.322	-0.019	0.028	-0.063	-0.049	-0.043	-0.045	-0.042	-0.049	-0.087	0.028
0.352	0.029	0.064	0.066	0.013	0.013	0.022	0.014	0.011	0.036	0.033
0.381	0.158	0.147		0.128	0.133	0.137	0.136	0.138	0.141	0.121
0.392	0.095	0.074	0.085	0.081	0.076	0.077	0.079	0.076	0.078	0.056
0.402	-0.010	-0.015	-0.020	-0.015	-0.012	-0.008	-0.011	-0.012	-0.010	-0.004
0.447	0.035	0.030	0.029	0.025	0.023		0.018	0.015	0.023	-0.004
0.492	0.032	0.029	0.027	0.024	0.016	0.017	0.018	0.020	0.024	0.018
0.529	0.023	0.024	0.021	-0.004	0.002	0.009	0.010	0.010	0.011	0.009
0.592	0.015	0.013	0.025	0.009	0.003	-0.000	0.001	0.003	0.004	-0.000
0.655	-0.116	-0.110	-0.116	-0.109	-0.110	-0.118	-0.084	-0.112	-0.107	-0.114
0.661	-0.141	-0.210	-0.134	-0.128	-0.120	-0.140		-0.130	-0.124	-0.114
0.687	-0.011	0.046	-0.019	-0.035	-0.026	-0.016	0.030	-0.026		-0.026

[redacted]

TABLE VIII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 0.70$ - Continued

(e) $\alpha = 3.9°$

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.231	0.871	0.857	0.820	0.755	0.731	0.599	0.597	0.632	0.688	0.741
0.234	0.568	0.544	0.516	0.453	0.414	0.300	0.305	0.333	0.383	0.424
0.246	0.232	0.216	0.166	0.105	0.081	0.004	0.003	0.024	0.056	0.096
0.251	0.057	0.046	0.035	-0.013	-0.062	-0.149	-0.131	-0.101	-0.085	-0.041
0.268	0.069	0.086	0.052	0.039	0.005	-0.018	-0.021	-0.000	-0.036	-0.010
0.286	-0.191	-0.200	-0.235	-0.276	-0.303	-0.355	-0.354	-0.319	-0.325	-0.309
0.292	-0.531	-0.548	-0.586	-0.646	-0.671	-0.681	-0.691	-0.808	-0.697	-0.669
0.322	-0.001	0.066	-0.091	-0.068	-0.061	-0.043	-0.041	-0.058	-0.126	-0.038
0.352	0.039	0.067	0.085	0.006	-0.004	0.031	0.023	0.011	0.040	-0.002
0.381	0.175	0.160		0.119	0.124	0.130	0.129	0.130	0.137	0.106
0.392	0.110	0.084	0.091	0.075	0.065	0.070	0.070	0.063	0.070	0.048
0.402	0.004	-0.006	-0.019	-0.028	-0.030	-0.010	-0.016	-0.026	-0.026	-0.023
0.447	0.057	0.049	0.030	0.012	0.002		0.008	-0.000	-0.003	-0.010
0.492	0.056	0.048	0.008	0.007	-0.008	0.010	0.008	0.005	-0.003	-0.003
0.529	0.050	0.046	0.023	0.008	0.002	0.003	0.004	0.001	-0.000	-0.000
0.592	0.028	0.025	0.014	0.003	-0.004	-0.005	-0.005	-0.003	-0.007	-0.008
0.655	-0.107	-0.107	-0.118	-0.117	-0.122	-0.114	-0.087	-0.113	-0.117	-0.126
0.661	-0.136	-0.210	-0.137	-0.137	-0.132	-0.132		-0.130	-0.134	-0.127
0.687	-0.004	0.055	-0.020	-0.043	-0.035	-0.016	0.023	-0.032		-0.036

(f) $\alpha = 6.0°$

0.231	0.926	0.908	0.854	0.761	0.714	0.512	0.510	0.564	0.648	0.725
0.234	0.629	0.600	0.550	0.455	0.394	0.227	0.230	0.271	0.342	0.405
0.246	0.287	0.264	0.197	0.107	0.068	-0.049	-0.051	-0.017	0.042	0.081
0.251	0.106	0.100	0.047	-0.018	-0.077	-0.211	-0.192	-0.165	-0.120	-0.036
0.268	0.103	0.116	0.068	0.035	-0.000	-0.048	-0.053	-0.032	-0.041	-0.016
0.286	-0.194	-0.204	-0.237	-0.282	-0.308	-0.361	-0.361	-0.326	-0.331	-0.316
0.292	-0.537	-0.549	-0.594	-0.654	-0.680	-0.689	-0.699	-0.818	-0.709	-0.681
0.322	-0.006	0.063	-0.100	-0.079	-0.071	-0.050	-0.048	-0.069	-0.149	-0.072
0.352	0.036	0.061	0.086	-0.010	-0.024	0.027	0.018	-0.000	0.033	-0.034
0.381	0.181	0.164		0.098	0.106	0.121	0.119	0.116	0.118	0.096
0.392	0.123	0.091	0.089	0.060	0.047	0.063	0.061	0.051	0.056	0.029
0.402	0.010	-0.004	-0.022	-0.040	-0.044	-0.014	-0.021	-0.033	-0.034	-0.032
0.447	0.058	0.048	0.025	0.005	-0.004		0.004	-0.004	-0.009	-0.016
0.492	0.054	0.046	0.023	0.004	-0.010	0.007	0.004	-0.000	-0.007	-0.008
0.529	0.047	0.042	0.008	-0.006	-0.015	-0.001	-0.003	-0.011	-0.019	-0.020
0.592	0.041	0.032	0.012	-0.011	-0.024	-0.012	-0.013	-0.019	-0.026	-0.027
0.655	-0.097	-0.109	-0.126	-0.134	-0.140	-0.115	-0.094	-0.121	-0.136	-0.146
0.661	-0.129	-0.211	-0.145	-0.155	-0.149	-0.132		-0.134	-0.154	-0.146
0.687	-0.000	0.058	-0.026	-0.059	0.037	-0.018	0.005	-0.054		-0.055

[redacted]

[redacted]

TABLE VIII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 0.70$ - Concluded

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	0.977	0.952	0.879	0.754	0.692	0.423	0.421	0.489	0.601	0.698
0.234	0.688	0.651	0.581	0.453	0.367	0.152	0.157	0.205	0.296	0.375
0.246	0.338	0.310	0.225	0.116	0.033	-0.106	-0.102	-0.068	-0.001	0.053
0.251	0.156	0.143	0.056	-0.033	-0.074	-0.287	-0.270	-0.221	-0.147	-0.059
0.268	0.139	0.146	0.078	0.013	-0.047	-0.082	-0.084	-0.068	-0.108	-0.068
0.286	-0.131	-0.148	-0.217	-0.303	-0.349	-0.405	-0.406	-0.379	-0.386	-0.360
0.292	-0.412	-0.443	-0.522	-0.645	-0.704	-0.659	-0.683	-0.836	-0.749	-0.705
0.322	0.017	0.111	-0.117	-0.127	-0.087	-0.047	-0.050	-0.092	-0.221	-0.227
0.352	0.049	0.061	0.094	-0.047	-0.068	0.028	0.015	-0.015	0.014	-0.038
0.381	0.194	0.169		0.062	0.078	0.110	0.107	0.101	0.091	0.068
0.392	0.136	0.097	0.082	0.038	0.017	0.056	0.051	0.034	0.022	-0.001
0.402	0.022		-0.025	-0.064	-0.073	-0.016	-0.026	-0.050	-0.069	-0.065
0.447	0.076	0.060	0.016	-0.022	-0.040		-0.006	-0.026	-0.045	-0.060
0.492	0.078	0.062	0.001	-0.036	-0.059	-0.002	-0.008	-0.025	-0.055	-0.059
0.529	0.077	0.061	0.014	-0.031	-0.043	-0.011	-0.014	-0.027	-0.047	-0.048
0.592	0.052	0.039	0.004	-0.034	-0.051	-0.020	-0.025	-0.033	-0.053	-0.056
0.655	-0.086	-0.113	-0.137	-0.158	-0.169	-0.120	-0.099	-0.134	-0.163	-0.174
0.661	-0.121	-0.214	-0.156	-0.181	-0.178	-0.139		-0.143	-0.178	-0.174
0.687	0.005	0.057	-0.036	-0.085	-0.078	-0.023	0.010	-0.047		-0.082

(h) $\alpha = 10.0°$

0.231	1.022	0.993	0.902	0.747	0.662	0.328	0.328	0.410	0.547	0.671
0.234	0.746	0.702	0.609	0.444	0.335	0.080	0.082	0.135	0.248	0.342
0.246	0.398	0.363	0.260	0.106	0.009	-0.154	-0.153	-0.117	-0.047	0.027
0.251	0.212	0.187	0.076	-0.032	-0.123	-0.375	-0.352	-0.280	-0.170	-0.089
0.268	0.179	0.179	0.094	0.012	-0.048	-0.111	-0.111	-0.105	-0.108	-0.069
0.286	-0.133	-0.155	-0.220	-0.301	-0.345	-0.400	-0.400	-0.368	-0.374	-0.349
0.292	-0.449	-0.465	-0.554	-0.658	-0.704	-0.680	-0.700	-0.840	-0.740	-0.704
0.322	-0.002	0.076	-0.121	-0.097	-0.089	-0.055	-0.054	-0.079	-0.159	-0.097
0.352	0.028	0.053	0.074	-0.013	-0.024	0.017	0.008	-0.004	0.025	-0.015
0.381	0.160	0.145		0.108	0.045	0.105	0.100	0.089	0.056	0.038
0.392	0.157	0.110	0.072	0.012	-0.012	0.052	0.042	0.002	-0.023	-0.038
0.402	0.040	0.011	-0.032	-0.090	-0.106	-0.017	-0.031	-0.067	-0.089	-0.095
0.447	0.090	0.068	0.013	-0.043	-0.061		-0.011	-0.037	-0.060	-0.077
0.492	0.085	0.065	0.012	-0.045	-0.068	-0.004	-0.011	-0.028	-0.059	-0.062
0.529	0.077	0.060	0.007	-0.057	-0.074	-0.016	-0.024	-0.038	-0.077	-0.081
0.592	0.070	0.052	-0.001	-0.059	-0.081	-0.024	-0.038	-0.044	-0.082	-0.087
0.655	-0.067	-0.113	-0.143	-0.182	-0.198	-0.122	-0.120	-0.139	-0.188	-0.203
0.661	-0.105	-0.212	-0.163	-0.206	-0.201	-0.148		-0.145	-0.202	-0.203
0.687	0.001	0.062	-0.045	-0.112	-0.107	-0.025	0.002	-0.052		-0.112

[redacted]

TABLE IX.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 0.90$

(a) $\alpha = -4.2°$

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.231	0.691	0.694	0.725	0.773	0.829	0.969	0.957	0.929	0.871	0.838
0.234	0.394	0.390	0.421	0.469	0.510	0.651	0.648	0.610	0.559	0.515
0.246	0.181	0.189	0.173	0.201	0.219	0.324	0.299	0.292	0.250	0.233
0.251	-0.563	-0.562	-0.466	-0.297	-0.271	0.009	0.005	-0.111	-0.174	-0.222
0.268	0.027	0.047	0.035	0.084	0.093	0.171	0.175	0.159	0.107	0.088
0.286	-0.111	-0.107	-0.103	-0.086	-0.073	-0.022	-0.025	-0.019	-0.056	-0.076
0.292	-1.005	-1.003	-0.999	-0.979	-0.986	-0.960	-0.955	-1.027	-0.987	-0.985
0.322	-0.017	-0.007	-0.005	0.002	0.011	0.028	0.030	0.029	0.042	0.262
0.352	0.057	0.111	0.062	0.046	0.044	0.048	0.044	0.037	0.035	0.059
0.381	0.167	0.170		0.169	0.161	0.202	0.200	0.193	0.175	0.146
0.392	0.090	0.074	0.081	0.087	0.092	0.139	0.138	0.121	0.101	0.069
0.402	-0.236	-0.031	-0.032	-0.037	-0.031	0.006	-0.002	-0.015	-0.023	-0.021
0.447	0.013	0.013	0.007	0.007	0.012		0.041	0.026	0.017	0.002
0.492	0.010	0.009	0.006	0.007	0.005	0.044	0.042	0.033	0.020	0.008
0.529	0.002	0.004	0.001	-0.000	0.002	0.033	0.033	0.022	0.008	-0.000
0.592	-0.007	-0.008	-0.008	-0.004	-0.003	0.021	0.020	0.015	0.004	-0.018
0.655	-0.172	-0.137	-0.168	-0.168	-0.166	-0.136	-0.018	-0.158	-0.169	-0.178
0.661	-0.223	-0.263	-0.204	-0.209	-0.188	-0.317		-0.367	-0.261	-0.182
0.687	-0.239	0.028	-0.044	-0.046	-0.039	-0.229	0.053	-0.024		-0.035

(b) $\alpha = -2.2°$

0.231	0.764	0.763	0.779	0.801	0.835	0.908	0.899	0.889	0.858	0.845
0.234	0.459	0.453	0.470	0.496	0.515	0.588	0.588	0.569	0.544	0.521
0.246	0.216	0.223	0.208	0.213	0.225	0.280	0.257	0.267	0.243	0.234
0.251	-0.365	-0.362	-0.325	-0.246	-0.268	-0.076	-0.093	-0.170	-0.261	-0.206
0.268	0.050	0.072	0.061	0.103	0.103	0.138	0.138	0.138	0.101	0.094
0.286	-0.082	-0.080	-0.080	-0.071	-0.068	-0.056	-0.056	-0.041	-0.063	-0.073
0.292	-0.999	-0.991	-0.989	-0.979	-0.980	-0.977	-0.969	-1.030	-0.990	-0.984
0.322	0.001	0.023	0.002	0.002	0.003	0.002	0.005	0.006	0.001	0.252
0.352	0.049	0.089	0.079	0.047	0.045	0.048	0.045	0.041	0.052	0.044
0.381	0.181	0.178		0.172	0.167	0.189	0.190	0.187	0.175	0.155
0.392	0.097	0.081	0.090	0.098	0.100	0.122	0.124	0.115	0.105	0.077
0.402	-0.024	-0.036	-0.028	-0.027	-0.025	-0.007	-0.012	-0.016	-0.015	-0.012
0.447	0.016	0.016	0.013	0.017	0.021		0.029	0.022	0.022	0.011
0.492	0.014	0.015	0.013	0.018	0.014	0.028	0.028	0.027	0.024	0.017
0.529	-0.007	0.008	0.006	0.007	0.008	0.014	0.016	0.012	0.008	0.004
0.592	-0.004	-0.005	-0.004	-0.001	-0.002	0.002	0.003	0.004	-0.000	-0.005
0.655	-0.177	-0.083	-0.175	-0.171	-0.168	-0.165	-0.029	-0.170	-0.170	-0.178
0.661	-0.290	-0.581	-0.218	-0.202	-0.189	-0.297		-0.317	-0.202	-0.183
0.687	-0.023	0.034	-0.029	-0.048	-0.041	-0.024	0.029	-0.036		-0.040

[redacted]

TABLE IX.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR M_∞=0.90 - Continued

(c) α =-0.1°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	0.840	0.835	0.834	0.829	0.841	0.849	0.840	0.845	0.843	0.851
0.234	0.528	0.521	0.523	0.524	0.524	0.527	0.533	0.532	0.529	0.531
0.246	0.262	0.266	0.244	0.236	0.240	0.243	0.227	0.246	0.244	0.247
0.251	-0.193	-0.182	-0.183	-0.174	-0.182	-0.180	-0.190	-0.193	-0.187	-0.165
0.268	0.089	0.110	0.095	0.114	0.109	0.115	0.113	0.123	0.105	0.101
0.286	-0.069	-0.069	-0.071	-0.065	-0.062	-0.050	-0.049	-0.032	-0.056	-0.067
0.292	-0.975	-0.977	-0.979	-0.970	-0.974	-0.969	-0.974	-1.025	-0.980	-0.976
0.322	-0.011	-0.000	-0.007	-0.000	0.006	0.024	0.027	0.026	0.037	0.255
0.352	0.060	0.115	0.061	0.043	0.040	0.047	0.043	0.032	0.026	0.075
0.381	0.159	0.161		0.162	0.175	0.182	0.183	0.185	0.181	0.170
0.392	0.113	0.095	0.109	0.110	0.110	0.114	0.117	0.113	0.111	0.091
0.402	-0.016	-0.025	-0.022	-0.014	-0.010	-0.004	-0.008	-0.009	-0.006	-0.000
0.447	0.031	0.030	0.029	0.029	0.031		0.029	0.026	0.030	0.022
0.492	0.027	0.026	0.026	0.029	0.024	0.029	0.031	0.032	0.031	0.027
0.529	0.015	0.017	0.016	0.015	0.015	0.018	0.020	0.018	0.015	0.011
0.592	0.003	0.002	0.010	0.006	0.005	0.011	0.013	0.013	0.011	0.001
0.655	-0.171	-0.081	-0.166	-0.162	-0.160	-0.146	-0.047	-0.159	-0.166	-0.173
0.661	-0.282	-0.709	-0.241	-0.193	-0.190	-0.295		-0.241	-0.219	-0.189
0.687	-0.024	0.022	-0.034	-0.051	-0.045	-0.010	0.056	-0.026		-0.042

(d) α = 1.9°

0.231	0.900	0.891	0.874	0.842	0.838	0.779	0.773	0.792	0.816	0.846
0.234	0.587	0.573	0.563	0.534	0.516	0.464	0.466	0.478	0.506	0.524
0.246	0.297	0.297	0.265	0.240	0.227	0.200	0.187	0.210	0.221	0.239
0.251	-0.080	-0.090	-0.186	-0.142	-0.194	-0.323	-0.336	-0.351	-0.239	-0.177
0.268	0.116	0.133	0.112	0.119	0.094	0.081	0.079	0.095	0.065	0.083
0.286	-0.017	-0.021	-0.040	-0.062	-0.078	-0.118	-0.114	-0.087	-0.091	-0.082
0.292	-0.974	-0.966	-0.975	-0.979	-0.985	-1.007	-0.996	-1.033	-1.004	-0.990
0.322	0.037	0.082	-0.003	-0.003	-0.006	-0.031	-0.029	-0.029	-0.048	0.236
0.352	0.046	0.079	0.106	0.037	0.032	0.057	0.051	0.043	0.074	0.028
0.381	0.196	0.185		0.167	0.168	0.166	0.167	0.170	0.181	0.159
0.392	0.121	0.098	0.112	0.106	0.101	0.098	0.100	0.098	0.105	0.079
0.402	-0.019	-0.028	-0.033	-0.027	-0.025	-0.015	-0.020	-0.023	-0.025	-0.015
0.447	0.034	0.030	0.026	0.018	0.015		0.010	0.007	0.015	-0.001
0.492	0.036	0.033	0.026	0.017	0.007	0.011	0.011	0.012	0.012	0.010
0.529	0.031	0.030	0.020	-0.009	0.001	0.001	0.001	-0.001	-0.003	-0.003
0.592	0.025	0.020	0.010	-0.000	-0.008	-0.011	-0.010	-0.008	-0.011	-0.012
0.655	-0.128	-0.076	-0.166	-0.183	-0.177	-0.171	-0.076	-0.171	-0.179	-0.189
0.661	-0.333	-0.759	-0.282	-0.236	-0.191	-0.237		-0.216	-0.205	-0.182
0.687	-0.011	0.051	-0.027	-0.057	-0.042	-0.029	0.005	-0.038		-0.051

[redacted]

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TABLE IX.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N-7 FOR $M_\infty = 0.90$ - Continued

(e) $\alpha = 3.9°$

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.231	0.963	0.950	0.916	0.856	0.833	0.711	0.707	0.739	0.790	0.841
0.234	0.653	0.635	0.604	0.550	0.511	0.401	0.404	0.433	0.480	0.519
0.246	0.343	0.337	0.287	0.244	0.231	0.167	0.157	0.180	0.212	0.240
0.251	0.022	0.003	-0.119	-0.151	-0.194	-0.536	-0.570	-0.521	-0.297	-0.282
0.268	0.152	0.166	0.136	0.120	0.085	0.056	0.053	0.073	0.047	0.071
0.286	0.012	0.007	-0.022	-0.060	-0.084	-0.135	-0.132	-0.105	-0.106	-0.090
0.292	-0.952	-0.947	-0.960	-0.970	-0.978	-1.007	-0.997	-1.024	-1.000	-0.990
0.322	0.051	0.113	-0.014	-0.020	-0.018	-0.056	-0.064	-0.057	-0.095	0.144
0.352	0.052	0.078	0.124	0.030	0.021	0.070	0.061	0.046	0.089	0.034
0.381	0.212	0.196		0.160	0.156	0.162	0.018	0.169	0.212	0.144
0.392	0.280	0.275	0.261	0.109	0.227	0.177	0.177	0.189	0.099	0.229
0.402	0.199	-0.016	0.180	-0.028	0.140	0.097	-0.010	0.108	-0.024	0.142
0.447	0.112	0.107	0.092	0.075	0.067		0.042	0.049	0.057	0.067
0.492	0.043	0.036	0.015	-0.006	-0.042	0.013	0.013	0.018	0.014	-0.081
0.529	0.059	0.037	0.049	0.016	0.015	0.020	0.004	0.077	0.036	0.062
0.592	0.025	0.017	0.012	-0.062	-0.092	-0.009	-0.009	-0.005	-0.115	-0.097
0.655	-0.123	-0.039	-0.165	-0.181	-0.191	-0.170	-0.094	-0.170	-0.180	-0.185
0.661	0.061	-0.751	-0.347	-0.242	-0.227	-0.054		-0.208	-0.200	-0.199
0.687	0.057	0.076	0.132	0.024	0.013	0.076	0.066	0.048		0.074

(f) $\alpha = 6.0°$

0.231	1.016	0.999	0.948	0.861	0.819	0.630	0.625	0.674	0.753	0.825
0.234	0.712	0.688	0.640	0.552	0.493	0.326	0.333	0.370	0.441	0.499
0.246	0.383	0.366	0.309	0.243	0.214	0.124	0.111	0.147	0.178	0.221
0.251	0.126	0.098	-0.000	-0.064	-0.202	-0.776	-0.756	-0.622	-0.358	-0.308
0.268	0.181	0.193	0.147	0.110	0.074	0.023	0.022	0.038	0.034	0.059
0.286	0.009	-0.281	-0.029	-0.069	-0.095	-0.147	-0.144	-0.116	-0.117	-0.100
0.292	-0.951	-0.956	-0.972	-0.984	-1.001	-1.025	-1.025	-1.038	-1.017	-1.006
0.322	0.043	0.107	-0.021	-0.026	-0.024	-0.062	-0.071	-0.065	-0.100	0.139
0.352	0.045	0.071	0.117	0.022	0.013	0.063	0.054	0.039	0.081	0.062
0.381	0.205	0.189		0.154	0.143	0.141	0.142	0.149	0.154	0.128
0.392	0.152	0.118	0.118	0.088	0.071	0.079	0.078	0.071	0.077	0.054
0.402	-0.000	-0.014	-0.034	-0.054	-0.058	-0.013	-0.022	-0.043	-0.045	-0.040
0.447	0.055	0.045	0.024	-0.005	-0.007		-0.009	-0.007	-0.012	-0.019
0.492	0.052	0.043	0.021	-0.000	-0.015	0.002	-0.011	-0.002	-0.011	-0.011
0.529	0.043	0.038	0.015	-0.012	0.066	-0.008	-0.009	0.066	-0.026	0.057
0.592	0.037	0.029	0.007	-0.017	-0.030	-0.021	-0.023	0.052	-0.033	-0.034
0.655	-0.107	0.050	-0.163	-0.201	-0.214	-0.171	-0.020	-0.182	-0.204	-0.212
0.661	-0.275	-0.674	-0.317	-0.207	-0.150	-0.115		-0.120	-0.139	-0.143
0.687	-0.003	0.058	-0.031	-0.072	-0.068	-0.031	0.008	-0.052		-0.072

[redacted]

[redacted]

TABLE IX.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 0.90$ - Concluded

$(\varepsilon)\ \alpha = 10.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	1.112	1.085	1.001	0.862	0.777	0.467	0.466	0.542	0.671	0.784
0.234	0.831	0.792	0.705	0.554	0.446	0.194	0.198	0.252	0.362	0.453
0.246	0.487	0.461	0.366	0.246	0.170	0.057	0.057	0.079	0.119	0.181
0.251	0.263	0.260	0.129	-0.090	-0.151	-0.146	-0.901	-0.860	-0.689	-0.171
0.268	0.262	0.260	0.177	0.103	0.057	-0.006	-0.003	-0.011	0.011	0.041
0.286	0.038	0.021	-0.018	-0.070	-0.104	-0.155	-0.151	-0.122	-0.123	-0.104
0.292	-0.912	-0.930	-0.958	-0.976	-1.017	-1.017	-1.036	-1.028	-1.014	-1.025
0.322	0.034	0.083	-0.027	-0.029	-0.094	-0.059	-0.059	-0.057	-0.076	-0.071
0.352	0.046	0.077	0.108	0.036	0.031	0.058	0.052	0.045	0.072	0.035
0.381	0.192	0.182		0.168	0.090	0.129	0.126	0.136	0.097	0.081
0.392	0.193	0.145	0.110	0.046	0.016	0.075	0.064	0.042	0.002	-0.008
0.402	0.038	0.009	-0.030	-0.093	-0.111	-0.003	-0.023	-0.067	-0.087	-0.087
0.447	0.093	0.073	0.021	-0.028	-0.045		-0.002	-0.023	-0.041	-0.055
0.492	0.081	0.064	0.021	-0.024	-0.043	0.002	-0.002	-0.014	-0.033	-0.034
0.529	0.064	0.053	0.012	-0.053	-0.071	-0.014	-0.022	-0.035	-0.073	-0.077
0.592	0.076	0.059	0.004	-0.056	-0.080	-0.028	-0.042	-0.047	-0.081	-0.086
0.655	-0.061	-0.022	-0.164	-0.231	-0.259	-0.168	-0.140	-0.197	-0.251	-0.257
0.661	-0.321	-0.744	-0.428	-0.437	-0.376	-0.214		-0.201	-0.280	-0.314
0.687	0.016	0.067	-0.044	-0.118	-0.115	-0.029	-0.000	-0.057		-0.122

[redacted]

[redacted]

TABLE X .- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.00$

(a) $\alpha = -4.2°$

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.231	0.778	0.780	0.810	0.855	0.906	1.039	1.026	0.999	0.945	0.915
0.234	0.490	0.489	0.518	0.564	0.600	0.733	0.728	0.693	0.646	0.603
0.246	0.313	0.303	0.301	0.328	0.343	0.434	0.415	0.403	0.369	0.356
0.251	-0.627	-0.604	-0.580	-0.551	-0.407	0.010	-0.024	-0.171	-0.361	-0.481
0.268	0.149	0.169	0.154	0.205	0.207	0.274	0.277	0.263	0.213	0.201
0.286	0.049	0.053	0.054	0.065	0.073	0.107	0.107	0.115	0.085	0.070
0.292	-0.648	-0.745	-0.741	-0.737	-0.690	-0.710	-0.712	-0.824	-0.729	-0.731
0.322	-0.309	-0.164	-0.171	-0.384	-0.285	-0.228	-0.346	-0.354	-0.168	0.039
0.352	-0.275	-0.275	-0.305	-0.278	-0.271	-0.275	-0.263	-0.275	-0.327	-0.410
0.381	0.147	0.155		0.151	0.148	0.173	0.178	0.172	0.142	0.115
0.392	0.101	0.091	0.092	0.103	0.114	0.161	0.163	0.146	0.118	0.085
0.402	-0.056	-0.073	-0.066	-0.098	-0.107	-0.071	-0.085	-0.098	-0.083	-0.064
0.447	0.035	0.038	0.030	0.029	0.034		0.056	0.045	0.039	0.028
0.492	0.059	0.060	0.056	0.058	0.056	0.082	0.081	0.076	0.069	0.060
0.529	0.052	0.056	0.053	0.052	0.054	0.071	0.073	0.065	0.057	0.051
0.592	0.045	0.046	0.045	0.050	0.051	0.067	0.067	0.064	0.056	0.046
0.655	-0.076	0.028	-0.075	-0.075	-0.067	-0.008	0.131	-0.028	-0.056	-0.082
0.661	-0.254	-0.559	-0.215	-0.229	-0.232	-0.227		-0.217	-0.239	-0.221
0.687	-0.128	-0.057	-0.155	-0.194	-0.166	-0.126	-0.039	-0.148		-0.174

(b) $\alpha = -2.2°$

0.231	0.848	0.847	0.861	0.881	0.913	0.988	0.973	0.964	0.936	0.923
0.234	0.553	0.549	0.564	0.590	0.608	0.676	0.676	0.657	0.635	0.612
0.246	0.346	0.339	0.337	0.342	0.352	0.399	0.384	0.387	0.366	0.359
0.251	-0.575	-0.557	-0.557	-0.534	-0.484	-0.239	-0.278	-0.321	-0.452	-0.508
0.268	0.170	0.190	0.180	0.212	0.212	0.247	0.247	0.247	0.214	0.205
0.286	0.059	0.062	0.062	0.072	0.076	0.098	0.098	0.110	0.085	0.074
0.292	-0.647	-0.737	-0.736	-0.734	-0.688	-0.718	-0.720	-0.829	-0.729	-0.730
0.322	-0.289	-0.137	-0.161	-0.368	-0.383	-0.254	-0.367	-0.361	-0.186	0.044
0.352	-0.273	-0.292	-0.291	-0.266	-0.267	-0.279	-0.269	-0.268	-0.299	-0.366
0.381	0.165	0.167		0.163	0.157	0.168	0.174	0.171	0.147	0.141
0.392	0.113	0.104	-0.065	0.118	0.123	0.146	0.150	0.141	0.125	0.102
0.402	-0.057	-0.077	-0.265	-0.072	-0.075	-0.065	-0.074	-0.075	-0.057	-0.052
0.447	0.044	0.046	0.040	0.042	0.047		0.055	0.050	0.049	0.040
0.492	0.064	0.066	0.063	0.067	0.064	0.079	0.080	0.078	0.073	0.066
0.529	0.056	0.060	0.058	0.059	0.061	0.071	0.072	0.069	0.063	0.058
0.592	0.049	0.050	0.049	0.053	0.054	0.063	0.064	0.063	0.058	0.051
0.655	-0.061	0.042	-0.063	-0.071	-0.069	-0.003	0.089	-0.038	-0.061	-0.075
0.661	-0.248	-0.544	-0.215	-0.221	-0.223	-0.237		-0.212	-0.227	-0.226
0.687	-0.128	-0.051	-0.147	-0.194	-0.168	-0.127	-0.052	-0.153		-0.163

[redacted]

[redacted]

TABLE X.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.00$ - Continued

(c) $\alpha = -0.1°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	0.917	0.913	0.911	0.906	0.919	0.926	0.917	0.922	0.919	0.926
0.234	0.617	0.609	0.613	0.613	0.614	0.617	0.621	0.619	0.621	0.618
0.246	0.382	0.388	0.366	0.359	0.364	0.365	0.350	0.367	0.366	0.369
0.251	-0.505	-0.495	-0.505	-0.508	-0.507	-0.508	-0.515	-0.501	-0.501	-0.503
0.268	0.199	0.217	0.204	0.225	0.218	0.222	0.221	0.230	0.210	0.209
0.286	0.084	0.084	0.081	0.083	0.083	0.082	0.083	0.100	0.084	0.079
0.292	-0.634	-0.723	-0.724	-0.725	-0.677	-0.724	-0.725	-0.835	-0.731	-0.723
0.322	-0.274	-0.120	-0.156	-0.359	-0.374	-0.270	-0.378	-0.367	-0.191	0.043
0.352	-0.275	-0.302	-0.293	-0.266	-0.268	-0.283	-0.270	-0.269	-0.302	-0.364
0.381	0.153	0.156		0.153	0.156	0.155	0.161	0.161	0.149	0.152
0.392	0.127	0.116	0.126	0.126	0.126	0.128	0.132	0.129	0.126	0.109
0.402	-0.080	-0.094	-0.082	-0.081	-0.078	-0.070	-0.067	-0.076	-0.073	-0.065
0.447	0.003	0.003	0.002	0.001	0.003		0.002	-0.000	0.003	-0.003
0.492	0.044	0.044	0.043	0.045	0.040	0.043	0.045	0.047	0.048	0.045
0.529	0.049	0.052	0.051	0.049	0.049	0.049	0.024	0.053	0.049	0.048
0.592	0.057	0.056	0.057	0.058	0.056	0.057	0.057	0.059	0.058	0.054
0.655	-0.041	0.053	-0.046	-0.066	-0.068	-0.037	0.076	-0.045	-0.061	-0.067
0.661	-0.240	-0.527	-0.214	-0.213	-0.214	-0.236		-0.205	-0.216	-0.226
0.687	-0.123	-0.046	-0.141	-0.187	-0.162	-0.123	-0.047	-0.148		-0.158

(d) $\alpha = 1.9°$

0.231	0.976	0.966	0.949	0.921	0.915	0.867	0.856	0.875	0.896	0.924
0.234	0.672	0.662	0.651	0.626	0.609	0.558	0.561	0.574	0.600	0.615
0.246	0.418	0.409	0.385	0.366	0.355	0.331	0.320	0.339	0.349	0.363
0.251	-0.258	-0.287	-0.351	-0.476	-0.515	-0.570	-0.582	-0.558	-0.528	-0.498
0.268	0.227	0.243	0.223	0.230	0.209	0.199	0.195	0.212	0.183	0.196
0.286	0.126	0.123	0.106	0.086	0.073	0.039	0.042	0.065	0.060	0.067
0.292	-0.639	-0.704	-0.712	-0.723	-0.683	-0.742	-0.742	-0.852	-0.739	-0.723
0.322	-0.193	-0.028	-0.129	-0.313	-0.381	-0.324	-0.421	-0.381	-0.244	0.036
0.352	-0.253	-0.317	-0.261	-0.254	-0.274	-0.263	-0.255	-0.263	-0.270	-0.381
0.381	0.175	0.162		0.158	0.159	0.152	0.157	0.158	0.168	0.148
0.392	0.147	0.129	0.143	0.132	0.126	0.118	0.121	0.121	0.135	0.107
0.402	-0.074	-0.081	-0.099	-0.071	-0.073	-0.048	-0.056	-0.065	-0.076	-0.057
0.447	0.054	0.050	0.048	0.040	0.040		0.037	0.033	0.040	0.028
0.492	0.079	0.076	0.070	0.064	0.055	0.059	0.059	0.060	0.061	0.057
0.529	0.078	0.078	0.070	0.057	0.053	0.053	0.054	0.051	0.049	0.048
0.592	0.080	0.075	0.066	0.056	0.048	0.044	0.044	0.046	0.044	0.043
0.655	0.014	0.095	-0.025	-0.057	-0.069	-0.069	0.003	-0.070	-0.079	-0.076
0.661	-0.215	-0.540	-0.226	-0.237	-0.228	-0.129		-0.216	-0.244	-0.217
0.687	-0.112	-0.028	-0.136	-0.206	-0.187	-0.130	-0.074	-0.161		-0.164

[redacted]

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TABLE X.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.00$ - Continued

(e) $\alpha = 3.9°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	288	315	342	270
0.231	1.026	1.012	0.980	0.927	0.902	0.793	0.784	0.817	0.863	0.911
0.234	0.725	0.708	0.680	0.629	0.593	0.488	0.492	0.519	0.565	0.599
0.246	0.445	0.433	0.392	0.355	0.344	0.288	0.278	0.299	0.326	0.351
0.251	0.017	-0.037	-0.155	-0.379	-0.523	-0.640	-0.647	-0.591	-0.560	-0.459
0.268	0.248	0.260	0.230	0.214	0.182	0.165	0.164	0.179	0.148	0.169
0.286	0.142	0.135	0.105	0.069	0.046	0.211	0.007	0.029	0.027	0.041
0.292	-0.654	-0.710	-0.723	-0.740	-0.700	-0.769	-0.771	-0.884	-0.763	-0.750
0.322	-0.152	0.006	-0.126	-0.303	-0.291	-0.359	-0.453	-0.403	-0.291	-0.004
0.352	-0.249	-0.337	-0.270	-0.271	-0.297	-0.236	-0.238	-0.270	-0.266	-0.445
0.381	0.172	0.149		0.146	0.145	0.137	0.140	0.141	0.162	0.113
0.392	0.155	0.130	0.139	0.122	0.107	0.099	0.100	0.099	0.116	0.079
0.402	-0.077	-0.281	-0.110	-0.111	-0.088	-0.031	-0.244	-0.065	-0.083	-0.055
0.447	0.073	0.065	0.054	0.039	0.033		0.038	0.032	0.031	0.022
0.492	0.092	0.086	0.068	0.052	0.039	0.054	0.052	0.050	0.043	0.042
0.529	0.087	0.083	0.063	0.038	0.029	0.041	0.040	0.033	0.024	0.024
0.592	0.085	0.076	0.055	0.032	0.020	0.026	0.025	0.020	0.016	0.014
0.655	0.024	0.095	-0.031	-0.075	-0.090	-0.114	-0.010	-0.085	-0.100	-0.098
0.661	-0.217	-0.560	-0.244	-0.255	-0.267	-0.086		-0.236	-0.260	-0.255
0.687	-0.124	-0.029	-0.148	-0.232	-0.202	-0.142	-0.085	-0.186		-0.180

(f) $\alpha = 6.0°$

0.231	1.081	1.064	1.016	0.936	0.894	0.724	0.716	0.763	0.835	0.902
0.234	0.788	0.765	0.721	0.639	0.583	0.429	0.435	0.472	0.537	0.589
0.246	0.487	0.469	0.419	0.363	0.340	0.262	0.250	0.280	0.309	0.339
0.251	0.162	0.069	-0.061	-0.241	-0.488	-0.668	-0.684	-0.644	-0.562	-0.469
0.268	0.286	0.293	0.252	0.209	0.167	0.153	0.153	0.160	0.126	0.151
0.286	0.171	0.162	0.119	0.067	0.035	-0.004	-0.002	-0.014	0.011	0.027
0.292	-0.644	-0.693	-0.709	-0.737	-0.691	-0.764	-0.768	-0.882	-0.766	-0.747
0.322	-0.089	0.068	-0.108	-0.279	-0.375	-0.369	-0.455	-0.404	-0.321	-0.046
0.352	-0.222	-0.332	-0.258	-0.282	-0.289	-0.196	-0.204	-0.261	-0.240	-0.492
0.381	0.157	0.130		0.130	0.138	0.137	0.132	0.134	0.149	0.099
0.392	0.178	0.144	0.144	0.121	0.098	0.099	0.096	0.094	0.110	0.083
0.402	-0.058	-0.058	-0.087	-0.150	-0.114	-0.013	-0.019	-0.053	-0.061	-0.047
0.447	0.095	0.085	0.063	0.039	0.032		0.046	0.037	0.026	0.018
0.492	0.111	0.102	0.075	0.048	0.031	0.060	0.056	0.050	0.036	0.032
0.529	0.108	0.100	0.068	0.029	0.017	0.044	0.040	0.027	0.012	0.028
0.592	0.108	0.094	0.060	0.023	0.006	0.029	0.022	0.016	0.002	0.019
0.655	0.049	0.106	-0.025	-0.080	-0.106	-0.094	-0.031	-0.115	-0.110	-0.132
0.661	-0.195	-0.548	-0.251	-0.276	-0.277	-0.108		-0.247	-0.284	-0.274
0.687	-0.078	-0.007	-0.127	-0.234	-0.236	-0.134	-0.092	-0.185		-0.208

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TABLE X.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.00$ - Continued

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	1.131	1.109	1.047	0.940	0.880	0.651	0.650	0.706	0.801	0.887
0.234	0.848	0.822	0.756	0.646	0.569	0.367	0.375	0.419	0.504	0.574
0.246	0.533	0.512	0.445	0.372	0.323	0.235	0.225	0.254	0.292	0.328
0.251	0.267	0.219	0.160	-0.044	-0.468	-0.697	-0.706	-0.665	-0.569	-0.460
0.268	0.324	0.327	0.269	0.191	0.135	0.146	0.149	0.140	0.088	0.118
0.286	0.206	0.191	0.130	0.055	0.017	-0.016	-0.018	-0.008	-0.013	0.008
0.292	-0.632	-0.671	-0.698	-0.736	-0.681	-0.761	-0.767	-0.885	-0.772	-0.754
0.322	-0.078	0.130	-0.092	-0.266	-0.359	-0.372	-0.448	-0.406	-0.354	-0.088
0.352	-0.191	-0.320	-0.228	-0.320	-0.290	-0.146	-0.164	-0.237	-0.232	-0.504
0.381	0.151	0.127		0.110	0.117	0.129	0.116	0.114	0.118	0.069
0.392	0.191	0.149	0.133	0.100	0.076	0.091	0.079	0.075	0.071	0.047
0.402	-0.072	-0.072	-0.089	-0.181	-0.212	-0.031	-0.041	-0.101	-0.112	-0.124
0.447	0.097	0.080	0.035	-0.007	-0.019		0.021	0.006	-0.016	-0.035
0.492	0.127	0.109	0.062	0.013	-0.005	0.053	0.040	0.026	0.003	-0.005
0.529	0.130	0.114	0.063	0.002	-0.013	0.040	0.029	0.008	-0.014	-0.020
0.592	0.133	0.114	0.062	0.004	-0.017	0.032	0.014	0.011	-0.018	-0.024
0.655	0.074	0.111	-0.025	-0.098	-0.127	-0.061	-0.046	-0.126	-0.128	-0.130
0.661	-0.174	-0.536	-0.251	-0.290	-0.298	-0.100		-0.251	-0.299	-0.287
0.687	-0.045	0.013	-0.112	-0.232	-0.251	-0.113	-0.445	-0.445		-0.231

(h) $\alpha = 10.0°$

0.231	1.169	1.144	1.064	0.933	0.854	0.564	0.564	0.633	0.755	0.860
0.234	0.898	0.864	0.780	0.637	0.537	0.293	0.301	0.352	0.454	0.542
0.246	0.572	0.548	0.464	0.358	0.292	0.194	0.184	0.212	0.247	0.299
0.251	0.337	0.331	0.199	-0.045	-0.466	-0.735	-0.745	-0.703	-0.591	-0.374
0.268	0.352	0.348	0.268	0.196	0.149	0.128	0.141	0.105	0.107	0.133
0.286	0.179	0.163	0.117	0.061	0.026	-0.007	-0.006	0.011	-0.028	0.026
0.292	-0.644	-0.686	-0.712	-0.744	-0.692	-0.770	-0.775	-0.889	-0.771	-0.758
0.322	-0.136	0.016	-0.123	-0.279	-0.385	-0.359	-0.450	-0.402	-0.343	-0.169
0.352	-0.249	-0.334	-0.269	-0.272	-0.319	-0.257	-0.253	-0.273	-0.275	-0.467
0.381	0.159	0.145		0.144	0.083	0.117	0.085	0.082	0.061	0.031
0.392	0.192	0.144	0.104	0.058	0.031	0.087	0.058	0.046	-0.001	0.003
0.402	-0.070	-0.085	-0.096	-0.189	-0.249	-0.041	-0.055	-0.109	-0.138	-0.178
0.447	0.094	0.073	0.020	-0.030	-0.041		0.008	-0.008	-0.032	-0.052
0.492	0.113	0.094	0.048	0.002	-0.015	0.039	0.029	0.017	-0.002	-0.008
0.529	0.106	0.094	0.054	0.005	-0.005	0.032	0.025	0.013	-0.004	-0.007
0.592	0.104	0.091	0.056	0.019	0.004	0.027	0.021	0.015	0.006	0.004
0.655	0.038	0.101	-0.027	-0.077	-0.102	-0.104	-0.060	-0.107	-0.101	-0.097
0.661	-0.208	-0.552	-0.277	-0.331	-0.340	-0.082		-0.261	-0.340	-0.330
0.687	-0.109	-0.029	-0.140	-0.245	-0.208	-0.137	-0.084	-0.173		-0.179

[redacted]

TABLE XI.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.10$

(a) $\alpha = -4.2°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	0.860	0.862	0.893	0.933	0.982	1.109	1.099	1.075	1.021	0.992
0.234	0.577	0.576	0.603	0.645	0.681	0.809	0.805	0.773	0.727	0.686
0.246	0.419	0.434	0.416	0.439	0.450	0.531	0.520	0.506	0.476	0.465
0.251	-0.460	-0.443	-0.446	-0.414	-0.371	-0.281	-0.291	-0.328	-0.345	-0.390
0.268	0.294	0.306	0.279	0.318	0.314	0.362	0.366	0.353	0.315	0.308
0.286	0.182	0.186	0.186	0.196	0.202	0.232	0.231	0.239	0.212	0.199
0.292	-0.549	-0.550	-0.548	-0.544	-0.542	-0.524	-0.526	-0.525	-0.542	-0.540
0.322	-0.223	-0.082	-0.055	-0.267	-0.260	-0.102	-0.228	-0.236	-0.052	0.138
0.352	-0.177	-0.201	-0.243	-0.199	-0.169	-0.162	-0.151	-0.166	-0.261	-0.320
0.381	0.179	0.180		0.177	0.148	0.125	0.140	0.153	0.139	0.114
0.392	0.170	0.163	0.157	0.170	0.180	0.218	0.223	0.209	0.179	0.149
0.402	-0.006	-0.022	-0.008	-0.027	-0.027	-0.001	-0.011	-0.021	-0.017	-0.000
0.447	0.022	0.025	0.017	0.019	0.023		0.047	0.035	0.030	0.017
0.492	0.021	0.021	0.018	0.018	0.014	0.039	0.039	0.034	0.027	0.019
0.529	0.028	0.032	0.031	0.031	0.033	0.047	0.049	0.043	0.035	0.029
0.592	0.031	0.031	0.029	0.032	0.032	0.051	0.052	0.050	0.042	0.034
0.655	0.006	0.075	0.001	-0.007	-0.006	0.099	0.190	0.053	0.017	-0.008
0.661	-0.145	-0.447	-0.115	-0.160	-0.161	-0.109		-0.108	-0.129	-0.127
0.687	-0.068	0.007	-0.097	-0.110	-0.074	-0.074	0.019	-0.103		-0.117

(b) $\alpha = -2.2°$

0.231	0.925	0.924	0.939	0.957	0.989	1.056	1.047	1.036	1.007	0.999
0.234	0.635	0.631	0.648	0.668	0.689	0.750	0.752	0.734	0.713	0.693
0.246	0.450	0.462	0.447	0.448	0.457	0.500	0.488	0.490	0.472	0.464
0.251	-0.431	-0.420	-0.423	-0.411	-0.384	-0.349	-0.347	-0.350	-0.358	-0.393
0.268	0.294	0.313	0.297	0.327	0.322	0.343	0.342	0.345	0.314	0.313
0.286	0.197	0.199	0.197	0.202	0.203	0.208	0.209	0.223	0.205	0.199
0.292	-0.546	-0.546	-0.545	-0.545	-0.544	-0.538	-0.540	-0.621	-0.548	-0.543
0.322	-0.176	-0.007	-0.047	-0.253	-0.265	-0.181	-0.267	-0.259	-0.090	0.128
0.352	-0.170	-0.229	-0.205	-0.164	-0.178	-0.187	-0.181	-0.173	-0.207	-0.296
0.381	0.153	0.140		0.160	0.160	0.148	0.162	0.166	0.146	0.136
0.392	0.179	0.173	0.174	0.184	0.189	0.206	0.212	0.204	0.186	0.167
0.402	-0.010	-0.029	-0.013	-0.017	-0.014	-0.001	-0.011	-0.013	-0.001	0.014
0.447	0.025	0.027	0.023	0.028	0.029		0.037	0.032	0.032	0.024
0.492	0.023	0.023	0.022	0.026	0.021	0.030	0.031	0.031	0.029	0.024
0.529	0.032	0.037	0.037	0.037	0.037	0.040	0.042	0.039	0.037	0.033
0.592	0.034	0.033	0.034	0.036	0.035	0.038	0.040	0.042	0.041	0.037
0.655	0.031	0.126	0.025	-0.000	-0.011	0.027	0.140	0.021	-0.008	-0.012
0.661	-0.132	-0.425	-0.122	-0.132	-0.136	-0.142		-0.114	-0.144	-0.137
0.687	-0.083	0.003	-0.103	-0.127	-0.093	-0.082	-0.013	-0.118		-0.088

[redacted]

TABLE XI.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR M_∞=1.10 - Continued

(c) α = -0.1°

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.231	0.989	0.983	0.983	0.976	0.990	0.997	0.989	0.994	0.990	0.999
0.234	0.692	0.684	0.690	0.688	0.692	0.692	0.697	0.694	0.700	0.696
0.246	0.479	0.489	0.468	0.464	0.464	0.467	0.467	0.470	0.466	0.470
0.251	-0.391	-0.384	-0.397	-0.399	-0.397	-0.401	-0.391	-0.396	-0.396	-0.393
0.268	0.302	0.324	0.311	0.328	0.322	0.327	0.327	0.334	0.315	0.313
0.286	0.196	0.195	0.194	0.200	0.202	0.216	0.217	0.230	0.208	0.198
0.292	-0.544	-0.543	-0.544	-0.544	-0.542	-0.535	-0.535	-0.618	-0.545	-0.541
0.322	-0.206	-0.055	-0.063	-0.272	-0.267	-0.138	-0.234	-0.242	-0.056	0.133
0.352	-0.187	-0.205	-0.238	-0.195	-0.180	-0.165	-0.152	-0.174	-0.253	-0.282
0.381	0.161	0.165		0.152	0.161	0.157	0.167	0.166	0.147	0.143
0.392	0.191	0.181	0.190	0.191	0.191	0.191	0.196	0.193	0.188	0.174
0.402	-0.013	-0.029	-0.026	-0.014	-0.012	-0.003	-0.010	-0.010	-0.007	-0.002
0.447	0.030	0.031	0.030	0.030	0.030		0.029	0.027	0.030	0.024
0.492	0.024	0.023	0.023	0.026	0.021	0.027	0.028	0.029	0.028	0.023
0.529	0.029	0.032	0.031	0.031	0.032	0.034	0.037	0.036	0.033	0.029
0.592	0.034	0.034	0.033	0.035	0.035	0.043	0.044	0.042	0.035	0.028
0.655	0.006	0.108	-0.001	-0.014	-0.015	0.062	0.180	0.045	0.010	-0.013
0.661	-0.153	-0.391	-0.127	-0.150	-0.156	-0.116		-0.110	-0.139	-0.141
0.687	-0.078	-0.013	-0.118	-0.128	-0.094	-0.084	0.016	-0.108		-0.113

(d) α = 1.9°

0.231	1.040	1.033	1.017	0.988	0.986	0.936	0.929	0.947	0.967	0.994
0.234	0.744	0.734	0.724	0.697	0.682	0.633	0.637	0.649	0.673	0.689
0.246	0.504	0.514	0.482	0.465	0.452	0.431	0.424	0.440	0.448	0.461
0.251	-0.357	-0.345	-0.358	-0.396	-0.408	-0.447	-0.439	-0.430	-0.417	-0.400
0.268	0.312	0.330	0.314	0.324	0.314	0.307	0.307	0.318	0.297	0.300
0.286	0.219	0.217	0.207	0.201	0.194	0.176	0.179	0.198	0.189	0.190
0.292	-0.543	-0.542	-0.546	-0.551	-0.554	-0.560	-0.561	-0.642	-0.562	-0.553
0.322	-0.157	0.012	-0.048	-0.248	-0.274	-0.208	-0.290	-0.276	-0.112	0.112
0.352	-0.171	-0.240	-0.205	-0.166	-0.186	-0.198	-0.191	-0.181	-0.211	-0.312
0.381	0.147	0.133		0.155	0.162	0.164	0.170	0.168	0.167	0.137
0.392	0.203	0.186	0.202	0.191	0.185	0.176	0.180	0.180	0.193	0.164
0.402	-0.017	-0.024	-0.050	-0.022	-0.019	-0.005	-0.010	-0.014	-0.042	0.007
0.447	0.037	0.033	0.032	0.027	0.025		0.020	0.018	0.027	0.018
0.492	0.032	0.030	0.027	0.025	0.018	0.020	0.020	0.020	0.022	0.017
0.529	0.040	0.041	0.037	0.030	0.027	0.025	0.027	0.026	0.026	0.025
0.592	0.038	0.035	0.033	0.030	0.025	0.023	0.025	0.029	0.030	0.027
0.655	0.054	0.143	0.032	0.004	-0.017	0.009	0.104	0.002	-0.016	-0.024
0.661	-0.126	-0.404	-0.126	-0.144	-0.143	-0.162		-0.129	-0.148	-0.145
0.687	-0.088	0.002	-0.109	-0.138	-0.105	-0.087	-0.022	-0.125		-0.095

[redacted]

TABLE XI.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.10$ - Continued

(e) $\alpha = 3.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	1.095	1.082	1.053	0.999	0.979	0.870	0.866	0.896	0.941	0.987
0.234	0.800	0.783	0.760	0.707	0.675	0.574	0.578	0.604	0.646	0.680
0.246	0.536	0.536	0.495	0.462	0.451	0.401	0.391	0.411	0.436	0.459
0.251	-0.293	-0.304	-0.352	-0.365	-0.411	-0.478	-0.470	-0.459	-0.431	-0.387
0.268	0.336	0.350	0.322	0.306	0.281	0.299	0.302	0.300	0.253	0.264
0.286	0.257	0.252	0.224	0.193	0.172	0.143	0.145	0.162	0.155	0.167
0.292	-0.523	-0.526	-0.537	-0.552	-0.561	-0.569	-0.573	-0.657	-0.571	-0.559
0.322	-0.077	0.101	-0.025	-0.211	-0.277	-0.242	-0.320	-0.290	-0.170	0.086
0.352	-0.138	-0.244	-0.194	-0.167	-0.187	-0.192	-0.186	-0.187	-0.197	-0.399
0.381	0.095	0.088		0.106	0.149	0.163	0.165	0.161	0.171	0.111
0.392	0.215	0.185	0.203	0.188	0.175	0.164	0.165	0.165	0.182	0.144
0.402	-0.021	-0.020	-0.051	-0.045	-0.035	-0.004	-0.009	-0.027	-0.048	-0.008
0.447	0.049	0.042	0.032	0.021	0.012		0.012	0.007	0.011	0.001
0.492	0.046	0.042	0.029	0.014	-0.000	0.019	0.017	0.011	0.004	-0.000
0.529	0.062	0.057	0.038	0.013	0.005	0.018	0.018	0.012	0.005	0.005
0.592	0.068	0.060	0.041	0.021	0.010	0.020	0.017	0.011	0.002	-0.000
0.655	0.112	0.185	0.056	0.006	-0.022	-0.027	0.053	-0.031	-0.031	-0.028
0.661	-0.094	-0.408	-0.132	-0.163	-0.170	-0.191		-0.136	-0.183	-0.174
0.687	-0.075	0.024	-0.097	-0.168	-0.146	-0.077	-0.034	-0.132		-0.117

(f) $\alpha = 6.0°$

0.231	1.149	1.131	1.086	1.009	0.971	0.805	0.804	0.846	0.912	0.978
0.234	0.859	0.838	0.797	0.716	0.667	0.505	0.523	0.559	0.620	0.672
0.246	0.574	0.563	0.523	0.467	0.444	0.377	0.363	0.393	0.416	0.461
0.251	0.027	-0.079	-0.270	-0.342	-0.399	-0.499	-0.491	-0.479	-0.437	-0.375
0.268	0.367	0.375	0.334	0.285	0.245	0.291	0.301	0.277	0.208	0.226
0.286	0.285	0.277	0.233	0.179	0.151	0.129	0.127	0.134	0.123	0.138
0.292	-0.502	-0.509	-0.526	-0.554	-0.568	-0.565	-0.574	-0.664	-0.581	-0.567
0.322	-0.068	0.191	-0.006	-0.186	-0.275	-0.256	-0.330	-0.299	-0.231	0.047
0.352	-0.096	-0.236	-0.156	-0.227	-0.180	-0.151	-0.154	-0.188	-0.163	-0.396
0.381	0.053	0.079		0.106	0.132	0.163	0.159	0.155	0.173	0.095
0.392	0.228	0.192	0.200	0.185	0.163	0.158	0.154	0.154	0.173	0.138
0.402	0.010	0.183	0.032	-0.003	-0.000	0.008	-0.003	-0.000	-0.005	-0.011
0.447	0.059	0.050	0.028	0.005	-0.012		0.014	0.005	-0.011	-0.016
0.492	0.081	0.074	0.043	0.003	-0.010	0.018	0.014	0.003	-0.015	-0.016
0.529	0.091	0.077	0.041	-0.000	-0.018	0.021	0.008	0.001	-0.025	-0.030
0.592	0.149	0.194	0.056	-0.012	-0.046	0.012	0.027	0.011	-0.059	-0.059
0.655	0.118	0.190	0.064	-0.199	-0.210	-0.022	0.057	-0.032	-0.220	-0.213
0.661	-0.086	-0.400	-0.125	-0.198	-0.199	-0.189		-0.135	0.012	-0.169
0.687	0.016	0.032	-0.090	-0.161	-0.142	-0.072	-0.028	-0.126		-0.108

[redacted]

TABLE XI.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE NO. FOR $M_\infty = 1.10$ - Concluded

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	40	45	72	90	180	200	225	252	270
0.231	1.190	1.171	1.110	1.008	0.950	0.728	0.726	0.784	0.875	0.959
0.234	0.910	0.882	0.823	0.715	0.642	0.444	0.454	0.500	0.579	0.649
0.246	0.606	0.591	0.531	0.467	0.422	0.340	0.329	0.358	0.391	0.437
0.251	0.297	0.167	0.043	-0.283	-0.384	-0.532	-0.525	-0.508	-0.438	-0.378
0.268	0.392	0.394	0.333	0.290	0.260	0.262	0.281	0.229	0.231	0.243
0.286	0.265	0.255	0.223	0.186	0.162	0.136	0.137	0.154	0.146	0.158
0.292	-0.517	-0.520	-0.537	-0.558	-0.571	-0.574	-0.580	-0.668	-0.580	-0.568
0.322	-0.076	0.101	-0.030	-0.215	-0.284	-0.251	-0.328	-0.297	-0.178	-0.015
0.352	-0.140	-0.251	-0.199	-0.174	-0.193	-0.197	-0.191	-0.192	-0.202	-0.411
0.381	0.087	0.083		0.096	0.101	0.157	0.142	0.136	0.118	0.065
0.392	0.231	0.187	0.175	0.159	0.137	0.147	0.132	0.130	0.120	0.100
0.402	-0.028	-0.037	-0.041	-0.115	-0.115	-0.006	-0.011	-0.080	-0.053	-0.061
0.447	0.069	0.055	0.023	-0.006	-0.019		-0.012	-0.010	-0.018	-0.026
0.492	0.054	0.043	0.017	-0.007	-0.022	0.009	0.005	-0.002	-0.016	-0.020
0.529	0.067	0.062	0.038	0.007	-0.013	0.013	0.010	0.001	-0.008	-0.008
0.592	0.063	0.053	0.029	0.004	-0.007	0.010	0.008	0.003	-0.004	-0.006
0.655	0.110	0.180	0.051	-0.000	-0.029	-0.031	0.046	-0.039	-0.037	-0.034
0.661	-0.096	-0.412	-0.137	-0.170	-0.177	-0.199		-0.142	-0.191	-0.181
0.687	-0.077	0.021	-0.101	-0.173	-0.154	-0.083	-0.039	-0.138		-0.120

(h) $\alpha = 10.0°$

0.231	1.237	1.211	1.137	1.011	0.935	0.657	0.659	0.726	0.841	0.941
0.234	0.969	0.934	0.857	0.716	0.625	0.379	0.399	0.448	0.545	0.631
0.246	0.652	0.634	0.559	0.468	0.403	0.316	0.304	0.332	0.364	0.424
0.251	0.396	0.346	0.252	-0.026	-0.375	-0.545	-0.538	-0.521	-0.416	-0.360
0.268	0.430	0.427	0.343	0.271	0.226	0.228	0.255	0.172	0.191	0.211
0.286	0.286	0.271	0.230	0.183	0.155	0.135	0.135	0.151	0.142	0.155
0.292	-0.502	-0.504	-0.528	-0.554	-0.569	-0.564	-0.571	-0.660	-0.577	-0.566
0.322	-0.084	0.086	-0.031	-0.220	-0.280	-0.243	-0.318	-0.292	-0.156	-0.072
0.352	-0.150	-0.249	-0.201	-0.170	-0.190	-0.200	-0.195	-0.187	-0.208	-0.322
0.381	0.110	0.099		0.121	0.056	0.153	0.123	0.117	0.059	0.042
0.392	0.239	0.190	0.146	0.126	0.109	0.151	0.116	0.109	0.052	0.064
0.402	-0.019	-0.049	-0.048	-0.121	-0.156	-0.000	-0.010	-0.096	-0.071	-0.115
0.447	0.091	0.071	0.023	-0.025	-0.043		-0.005	-0.019	-0.036	-0.049
0.492	0.073	0.056	0.014	-0.030	-0.046	0.009	0.001	-0.010	-0.035	-0.039
0.529	0.078	0.068	0.033	-0.007	-0.018	0.010	0.005	-0.004	-0.019	-0.020
0.592	0.078	0.067	0.038	-0.029	-0.007	0.013	0.004	0.002	-0.010	-0.011
0.655	0.110	0.181	0.050	-0.001	-0.030	-0.024	0.053	-0.031	-0.034	-0.031
0.661	-0.097	-0.405	-0.131	-0.162	-0.171	-0.184		-0.139	-0.172	-0.168
0.687	-0.089	-0.006	-0.113	-0.163	-0.133	-0.088	-0.032	-0.131		-0.102

[redacted]

[redacted]

TABLE XII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR M_∞=1.20

(a) α = -4.2°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	0.915	0.917	0.947	0.990	1.041	1.164	1.153	1.130	1.075	1.047
0.234	0.631	0.625	0.656	0.699	0.735	0.841	0.856	0.823	0.775	0.736
0.246	0.502	0.515	0.502	0.524	0.534	0.612	0.600	0.585	0.555	0.551
0.251	-0.330	-0.314	-0.319	-0.295	-0.271	-0.194	-0.191	-0.210	-0.250	-0.275
0.268	0.117	0.153	0.176	0.260	0.274	0.366	0.372	0.349	0.286	0.267
0.286	0.246	0.248	0.248	0.258	0.265	0.299	0.298	0.303	0.274	0.261
0.292	-0.405	-0.405	-0.405	-0.403	-0.403	-0.385	-0.388	-0.454	-0.403	-0.401
0.322	-0.170	-0.008	-0.013	-0.210	-0.201	-0.124	-0.179	-0.185	-0.021	0.159
0.352	-0.134	-0.190	-0.195	-0.139	-0.132	-0.125	-0.109	-0.124	-0.194	-0.272
0.381	0.094	0.056		0.077	0.038	0.002	-0.021	0.019	0.046	0.027
0.392	0.174	0.170	0.158	0.173	0.182	0.202	0.214	0.205	0.172	0.143
0.402	0.020	-0.038	0.024	-0.002	0.001	0.020	0.017	0.011	0.005	0.023
0.447	0.023	0.026	0.018	0.021	0.023		0.047	0.035	0.029	0.016
0.492	0.017	0.018	0.015	0.017	0.015	0.042	0.041	0.035	0.026	0.016
0.529	0.024	0.027	0.018	0.016	0.019	0.054	0.054	0.042	0.026	0.016
0.592	0.023	0.026	0.020	0.019	0.021	0.055	0.053	0.045	0.030	0.020
0.655	0.024	0.108	0.007	0.002	0.002	0.112	0.216	0.074	0.022	-0.000
0.661	-0.102	-0.362	-0.092	-0.128	-0.129	-0.047		-0.061	-0.111	-0.120
0.687	-0.069	-0.002	-0.112	-0.096	-0.072	-0.085	0.035	-0.108		-0.089

(b) α = -2.2°

0.231	0.985	0.984	1.000	1.018	1.050	1.115	1.105	1.094	1.065	1.056
0.234	0.687	0.686	0.705	0.727	0.747	0.788	0.809	0.789	0.766	0.747
0.246	0.536	0.548	0.536	0.538	0.546	0.588	0.575	0.576	0.558	0.559
0.251	-0.298	-0.286	-0.293	-0.279	-0.268	-0.231	-0.227	-0.242	-0.258	-0.265
0.268	0.186	0.207	0.218	0.283	0.286	0.328	0.331	0.325	0.285	0.279
0.286	0.266	0.268	0.267	0.273	0.274	0.281	0.281	0.293	0.276	0.270
0.292	-0.393	-0.393	-0.393	-0.392	-0.392	-0.386	-0.389	-0.451	-0.396	-0.391
0.322	-0.145	0.005	-0.000	-0.194	-0.194	-0.146	-0.197	-0.194	-0.045	0.161
0.352	-0.118	-0.203	-0.180	-0.111	-0.129	-0.134	-0.098	-0.122	-0.179	-0.265
0.381	0.038	0.018		0.048	0.061	0.032	0.030	0.051	0.061	0.022
0.392	0.182	0.176	0.173	0.188	0.193	0.198	0.209	0.203	0.181	0.163
0.402	0.032	-0.025	0.016	0.011	0.015	0.023	0.019	0.017	0.024	0.031
0.447	0.030	0.031	0.027	0.032	0.033		0.038	0.032	0.033	0.024
0.492	0.023	0.025	0.023	0.028	0.025	0.035	0.036	0.036	0.033	0.027
0.529	0.032	0.035	0.033	0.033	0.035	0.040	0.043	0.040	0.036	0.033
0.592	0.036	0.036	0.037	0.039	0.037	0.035	0.036	0.039	0.039	0.035
0.655	0.060	0.157	0.049	0.017	0.009	0.058	0.171	0.049	0.021	0.010
0.661	-0.077	-0.313	-0.074	-0.106	-0.112	-0.086		-0.067	-0.112	-0.109
0.687	-0.084	0.017	-0.106	-0.095	-0.064	-0.079	0.001	-0.119		-0.061

[redacted]

[redacted]

TABLE XII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.20$ - Continued

(c) $\alpha = -0.1°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	1.044	1.039	1.037	1.032	1.048	1.052	1.045	1.049	1.043	1.054
0.234	0.746	0.738	0.744	0.742	0.745	0.727	0.749	0.746	0.749	0.746
0.246	0.557	0.567	0.549	0.544	0.546	0.548	0.540	0.549	0.548	0.558
0.251	-0.277	-0.269	-0.281	-0.282	-0.282	-0.282	-0.274	-0.279	-0.281	-0.279
0.268	0.280	0.298	0.285	0.286	0.288	0.304	0.304	0.308	0.289	0.281
0.286	0.255	0.254	0.253	0.260	0.265	0.287	0.287	0.296	0.270	0.258
0.292	-0.408	-0.407	-0.409	-0.409	-0.408	-0.398	-0.400	-0.464	-0.411	-0.408
0.322	-0.172	-0.021	-0.021	-0.218	-0.209	-0.121	-0.178	-0.188	-0.019	0.145
0.352	-0.142	-0.186	-0.181	-0.153	-0.136	-0.125	-0.103	-0.126	-0.213	-0.256
0.381	0.105	0.079		0.073	0.078	0.073	0.083	0.083	0.068	0.025
0.392	0.191	0.179	0.004	0.191	0.194	0.189	0.197	0.194	0.183	0.171
0.402	0.011	-0.014	-0.004	0.010	0.013	0.021	0.015	0.014	0.017	0.011
0.447	0.028	0.028	0.028	0.032	0.032		0.031	0.027	0.029	0.021
0.492	0.018	0.019	0.017	0.021	0.016	0.025	0.027	0.026	0.024	0.019
0.529	0.024	0.027	0.032	0.030	0.030	0.029	0.032	0.032	0.031	0.030
0.592	0.034	0.034	0.035	0.036	0.034	0.035	0.035	0.037	0.035	0.031
0.655	0.058	0.155	0.047	0.015	0.007	0.059	0.164	0.047	0.016	0.010
0.661	-0.087	-0.315	-0.075	-0.109	-0.115	-0.086		-0.069	-0.109	-0.119
0.687	-0.082	0.007	-0.113	-0.097	-0.064	-0.080	0.009	-0.116		-0.062

(d) $\alpha = 1.9°$

0.231	1.101	1.092	1.079	1.051	1.049	0.996	0.991	1.008	1.025	1.055
0.234	0.802	0.790	0.784	0.757	0.743	0.669	0.695	0.707	0.728	0.744
0.246	0.590	0.599	0.571	0.555	0.546	0.525	0.515	0.533	0.538	0.562
0.251	-0.234	-0.233	-0.250	-0.264	-0.277	-0.308	-0.300	-0.296	-0.284	-0.273
0.268	0.305	0.320	0.298	0.299	0.270	0.240	0.236	0.267	0.237	0.259
0.286	0.311	0.308	0.290	0.271	0.258	0.236	0.237	0.252	0.245	0.252
0.292	-0.385	-0.388	-0.392	-0.400	-0.404	-0.406	-0.408	-0.473	-0.411	-0.404
0.322	-0.113	0.090	0.001	-0.190	-0.207	-0.173	-0.226	-0.218	-0.087	0.154
0.352	-0.107	-0.218	-0.178	-0.110	-0.131	-0.147	-0.152	-0.128	-0.188	-0.274
0.381	0.001	0.012		0.004	0.069	0.095	0.105	0.094	0.047	0.017
0.392	0.201	0.181	0.197	0.193	0.192	0.180	0.186	0.185	0.195	0.165
0.402	0.012	-0.000	-0.034	0.011	0.011	0.023	0.019	0.018	-0.025	0.026
0.447	0.044	0.041	0.037	0.031	0.026		0.022	0.019	0.028	0.019
0.492	0.039	0.036	0.029	0.024	0.014	0.017	0.017	0.017	0.018	0.004
0.529	0.050	0.049	0.039	0.025	0.021	0.022	0.023	0.021	0.020	0.020
0.592	0.053	0.050	0.042	0.032	0.022	0.022	0.021	0.024	0.020	0.019
0.655	0.109	0.193	0.067	0.018	-0.000	0.027	0.142	0.023	-0.000	-0.001
0.661	-0.049	-0.311	-0.072	-0.113	-0.127	-0.117		-0.096	-0.128	-0.122
0.687	-0.090	0.030	-0.102	-0.115	-0.086	-0.079	-0.020	-0.128		-0.072

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TABLE XII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.20$ - Continued

(e) $\alpha = 3.9°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	1.153	1.142	1.113	1.059	1.041	0.931	0.927	0.957	0.999	1.047
0.234	0.856	0.839	0.817	0.765	0.734	0.610	0.635	0.661	0.700	0.736
0.246	0.617	0.622	0.589	0.552	0.542	0.496	0.484	0.502	0.524	0.555
0.251	-0.197	-0.196	-0.225	-0.257	-0.282	-0.336	-0.329	-0.324	-0.300	-0.275
0.268	0.343	0.357	0.321	0.297	0.253	0.169	0.147	0.212	0.205	0.240
0.286	0.331	0.325	0.295	0.259	0.239	0.218	0.217	0.228	0.221	0.231
0.292	-0.379	-0.383	-0.392	-0.406	-0.414	-0.412	-0.417	-0.484	-0.423	-0.414
0.322	-0.086	0.130	0.010	-0.188	-0.218	-0.183	-0.238	-0.229	-0.118	0.128
0.352	-0.094	-0.221	-0.169	-0.126	-0.129	-0.150	-0.156	-0.134	-0.192	-0.308
0.381	-0.014	0.017		-0.054	0.046	0.113	0.117	0.097	-0.002	0.014
0.392	0.205	0.177	0.191	0.185	0.180	0.171	0.172	0.173	0.188	0.144
0.402	0.009	0.003	-0.026	-0.002	-0.000	0.025	0.021	0.010	-0.030	0.020
0.447	0.053	0.047	0.038	0.026	0.018		0.018	0.013	0.017	0.007
0.492	0.049	0.043	0.029	0.005	0.002	0.015	0.013	0.010	0.004	0.003
0.529	0.063	0.059	0.040	0.005	0.008	0.017	0.017	0.013	0.007	0.007
0.592	0.067	0.060	0.042	0.019	0.008	0.019	0.017	0.012	0.003	0.002
0.655	0.135	0.209	0.076	0.014	-0.012	0.014	0.108	-0.003	-0.011	-0.012
0.661	-0.030	-0.308	-0.071	-0.120	-0.137	-0.130		-0.092	-0.143	-0.136
0.687	-0.088	0.038	-0.097	-0.133	-0.109	-0.074	-0.021	-0.121		-0.089

(f) $\alpha = 6.0°$

0.231	1.203	1.187	1.145	1.066	1.030	0.864	0.861	0.903	0.969	1.036
0.234	0.910	0.889	0.850	0.771	0.722	0.548	0.575	0.611	0.670	0.722
0.246	0.647	0.638	0.602	0.552	0.530	0.465	0.453	0.480	0.504	0.545
0.251	-0.151	-0.148	-0.190	-0.247	-0.283	-0.359	-0.351	-0.344	-0.312	-0.269
0.268	0.381	0.390	0.343	0.296	0.248	0.123	0.094	0.165	0.195	0.233
0.286	0.338	0.329	0.296	0.255	0.233	0.214	0.213	0.224	0.216	0.226
0.292	-0.375	-0.379	-0.392	-0.409	-0.418	-0.414	-0.419	-0.489	-0.428	-0.418
0.322	-0.082	0.133	0.007	-0.187	-0.219	-0.185	-0.239	-0.231	-0.121	0.074
0.352	-0.093	-0.221	-0.167	-0.127	-0.129	-0.150	-0.156	-0.135	-0.193	-0.359
0.381	-0.015	0.018		-0.057	0.007	0.124	0.122	0.100	-0.021	0.014
0.392	0.213	0.178	0.182	0.173	0.165	0.165	0.161	0.161	0.171	0.127
0.402	0.010	0.001	-0.010	-0.033	-0.020	0.027	0.022	-0.008	0.003	0.006
0.447	0.065	0.055	0.035	0.017	0.008		0.015	0.007	0.002	-0.003
0.492	0.058	0.050	0.028	0.009	-0.006	0.014	0.011	0.006	-0.002	-0.003
0.529	0.068	0.063	0.041	0.012	0.003	0.017	0.015	0.010	0.002	0.003
0.592	0.069	0.062	0.042	0.018	0.004	0.035	0.016	0.009	-0.000	-0.000
0.655	0.137	0.210	0.077	0.014	-0.013	0.014	0.085	-0.005	-0.012	-0.013
0.661	-0.028	-0.307	-0.071	-0.121	-0.138	-0.131		-0.092	-0.144	-0.137
0.687	-0.087	0.039	-0.096	-0.134	-0.111	-0.073	-0.021	-0.121		-0.090

[redacted]

TABLE XII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.20$ - Concluded

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	0	45	72	90	180	200	225	252	270
0.231	1.248	1.229	1.171	1.069	1.015	0.792	0.791	0.844	0.935	1.019
0.234	0.964	0.937	0.880	0.773	0.702	0.484	0.514	0.557	0.633	0.703
0.246	0.678	0.665	0.611	0.552	0.514	0.436	0.423	0.450	0.481	0.524
0.251	-0.100	-0.136	-0.154	-0.237	-0.284	-0.380	-0.372	-0.364	-0.327	-0.264
0.268	0.419	0.423	0.359	0.287	0.224	0.085	0.093	0.120	0.154	0.205
0.286	0.362	0.349	0.299	0.230	0.190	0.195	0.192	0.196	0.177	0.181
0.292	-0.365	-0.371	-0.392	-0.424	-0.435	-0.420	-0.427	-0.506	-0.443	-0.435
0.322	-0.045	0.177	0.011	-0.181	-0.233	-0.192	-0.248	-0.241	-0.160	0.020
0.352	-0.078	-0.220	-0.149	-0.162	-0.140	-0.148	-0.152	-0.145	-0.169	-0.363
0.381	-0.013	0.017		-0.112	-0.051	0.133	0.122	0.102	-0.008	0.004
0.392	0.219	0.184	0.167	0.142	0.142	0.163	0.148	0.143	0.123	0.098
0.402	0.015	-0.005	-0.014	-0.059	-0.056	0.026	0.027	-0.044	-0.025	-0.036
0.447	0.079	0.064	0.029	0.002	-0.013		0.008	-0.001	-0.013	-0.025
0.492	0.076	0.062	0.027	-0.010	-0.029	0.012	0.006	-0.004	-0.022	-0.023
0.529	0.088	0.078	0.042	-0.002	-0.014	0.016	0.010	-0.000	-0.017	-0.016
0.592	0.088	0.076	0.041	0.001	-0.016	0.017	0.009	0.001	-0.020	-0.022
0.655	0.164	0.219	0.082	0.006	-0.028	0.042	0.071	-0.020	-0.030	-0.032
0.661	-0.010	-0.302	-0.075	-0.135	-0.152	-0.126		-0.106	-0.163	-0.157
0.687	-0.080	0.046	-0.088	-0.155	-0.143	-0.064	-0.023	-0.119		-0.113

(h) $\alpha = 10.0°$

0.231	1.288	1.264	1.190	1.064	0.992	0.715	0.715	0.781	0.895	0.998
0.234	1.014	0.981	0.905	0.769	0.676	0.417	0.445	0.496	0.590	0.677
0.246	0.708	0.696	0.623	0.543	0.483	0.403	0.390	0.416	0.444	0.500
0.251	0.297	0.162	-0.071	-0.231	-0.284	-0.404	-0.398	-0.387	-0.337	-0.261
0.268	0.459	0.458	0.375	0.271	0.189	0.038	0.096	0.064	0.106	0.169
0.286	0.385	0.367	0.299	0.203	0.141	0.165	0.160	0.152	0.099	0.133
0.292	-0.354	-0.363	-0.394	-0.440	-0.465	-0.431	-0.442	-0.705	-0.469	-0.464
0.322	-0.003	0.222	0.010	-0.184	-0.252	-0.199	-0.261	-0.255	-0.205	-0.042
0.352	-0.068	-0.224	-0.125	-0.205	-0.174	-0.144	-0.149	-0.160	-0.130	-0.261
0.381	-0.001	0.007		-0.136	-0.152	0.131	0.106	0.092	-0.019	-0.032
0.392	0.229	0.192	0.150	0.085	0.104	0.160	0.123	0.114	0.051	0.059
0.402	0.007	-0.012	-0.031	-0.094	-0.100	0.027	0.010	-0.084	-0.044	-0.086
0.447	0.094	0.073	0.019	-0.027	-0.044		-0.006	-0.019	-0.041	-0.059
0.492	0.092	0.071	0.020	-0.038	-0.062	0.005	-0.009	-0.019	-0.052	-0.057
0.529	0.103	0.087	0.034	-0.032	-0.049	0.007	-0.006	-0.013	-0.047	-0.051
0.592	0.104	0.086	0.033	-0.026	-0.050	0.003	-0.010	-0.014	-0.051	-0.057
0.655	0.185	0.219	0.075	-0.017	-0.058	0.065	0.058	-0.040	-0.059	-0.065
0.661	0.003	-0.302	-0.086	-0.158	-0.179	-0.106		-0.117	-0.191	-0.188
0.687	-0.072	0.049	-0.087	-0.184	-0.182	-0.070	-0.017	-0.121		-0.148

TABLE XIII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR M_∞=1.28

(a) α = -4.2°

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.231	0.940	0.942	0.973	1.018	1.067	1.198	1.183	1.158	1.104	1.079
0.234	0.648	0.643	0.674	0.720	0.755	0.881	0.875	0.846	0.800	0.756
0.246	0.556	0.570	0.556	0.578	0.588	0.663	0.660	0.635	0.608	0.602
0.251	-0.241	-0.229	-0.232	-0.211	-0.193	-0.123	-0.140	-0.142	-0.175	-0.192
0.268	0.075	0.100	0.120	0.201	0.225	0.334	0.340	0.315	0.236	0.207
0.286	0.243	0.227	0.229	0.244	0.257	0.308	0.304	0.302	0.266	0.248
0.292	-0.331	-0.331	-0.330	-0.326	-0.323	-0.297	-0.302	-0.355	-0.320	-0.322
0.322	-0.143	0.008	0.006	-0.176	-0.165	-0.108	-0.140	-0.147	-0.002	0.169
0.352	-0.112	-0.179	-0.182	-0.111	-0.103	-0.106	-0.078	-0.092	-0.178	-0.279
0.381	0.031	-0.026		-0.002	-0.051	-0.025	-0.060	-0.060	-0.004	-0.011
0.392	0.156	0.150	0.138	0.154	0.158	0.180	0.189	0.185	0.151	0.122
0.402	0.020	-0.056	0.011	0.001	0.008	0.030	0.032	0.023	-0.000	0.023
0.447	0.021	0.022	0.015	0.014	0.018		0.042	0.029	0.023	0.011
0.492	0.010	0.013	0.008	0.010	0.008	0.041	0.040	0.034	0.022	0.010
0.529	0.019	0.022	0.019	0.019	0.022	0.045	0.046	0.035	0.023	0.016
0.592	0.021	0.022	0.022	0.023	0.024	0.041	0.040	0.036	0.026	0.017
0.655	0.044	0.116	0.030	0.011	0.013	0.093	0.224	0.063	0.025	0.010
0.661	-0.070	-0.287	-0.066	-0.105	-0.105	-0.036		-0.043	-0.097	-0.097
0.687	-0.077	0.008	-0.111	-0.079	-0.061	-0.093	0.028	-0.112		-0.055

(b) α = -2.2°

0.231	1.011	1.010	1.024	1.046	1.077	1.141	1.132	1.121	1.093	1.087
0.234	0.710	0.705	0.722	0.746	0.766	0.816	0.823	0.810	0.791	0.766
0.246	0.586	0.599	0.586	0.591	0.606	0.635	0.626	0.627	0.612	0.611
0.251	-0.214	-0.208	-0.212	-0.199	-0.192	-0.159	-0.172	-0.168	-0.182	-0.188
0.268	0.140	0.160	0.160	0.215	0.216	0.277	0.279	0.270	0.208	0.203
0.286	0.276	0.261	0.260	0.266	0.268	0.272	0.271	0.281	0.265	0.259
0.292	-0.320	-0.320	-0.318	-0.316	-0.315	-0.310	-0.312	-0.363	-0.318	-0.315
0.322	-0.133	0.040	0.016	-0.163	-0.159	-0.122	-0.153	-0.154	-0.015	0.176
0.352	-0.104	-0.189	-0.172	-0.096	-0.102	-0.110	-0.095	-0.097	-0.168	-0.235
0.381	0.002	-0.027		-0.008	-0.020	-0.016	-0.042	-0.025	0.017	-0.024
0.392	0.160	0.150	0.147	0.165	0.169	0.171	0.184	0.181	0.157	0.138
0.402	0.017	-0.037	0.009	0.010	0.017	0.027	0.027	0.024	0.024	0.025
0.447	0.023	0.024	0.012	0.025	0.028		0.030	0.027	0.029	0.011
0.492	0.018	0.021	0.020	0.032	0.018	0.026	0.028	0.029	0.025	0.020
0.529	0.027	0.031	0.030	0.030	0.031	0.033	0.035	0.031	0.027	0.025
0.592	0.031	0.031	0.033	0.034	0.033	0.032	0.032	0.034	0.030	0.025
0.655	0.067	0.162	0.052	0.015	0.017	0.070	0.181	0.055	0.025	0.020
0.661	-0.058	-0.261	-0.055	-0.098	-0.100	-0.055		-0.046	-0.095	-0.094
0.687	-0.084	0.014	-0.110	-0.075	-0.054	-0.083	0.016	-0.112		-0.048

[redacted]

TABLE XIII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N-7 FOR M_∞=1.28 - Continued

(c) α =-0.1°

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	1.073	1.069	1.068	1.067	1.080	1.090	1.077	1.080	1.076	1.086
0.234	0.769	0.760	0.766	0.770	0.770	0.768	0.766	0.771	0.778	0.769
0.246	0.617	0.617	0.607	0.605	0.608	0.605	0.608	0.607	0.607	0.617
0.251	-0.187	-0.178	-0.190	-0.187	-0.190	-0.191	-0.206	-0.189	-0.190	-0.187
0.268	0.202	0.217	0.205	0.222	0.219	0.218	0.219	0.223	0.202	0.203
0.286	0.284	0.268	0.267	0.271	0.272	0.270	0.272	0.286	0.273	0.267
0.292	-0.314	-0.313	-0.313	-0.313	-0.313	-0.312	-0.313	-0.359	-0.315	-0.312
0.322	-0.130	0.044	0.015	-0.161	-0.159	-0.123	-0.155	-0.156	-0.016	0.176
0.352	-0.103	-0.191	-0.171	-0.093	-0.101	-0.112	-0.098	-0.099	-0.167	-0.233
0.381	0.005	-0.023		-0.006	-0.006	0.021	0.003	0.010	0.022	-0.032
0.392	0.167	0.152	0.158	0.169	0.174	0.169	0.181	0.178	0.163	0.149
0.402	0.016	-0.015	-0.002	0.016	0.021	0.030	0.027	0.025	0.025	0.014
0.447	0.031	0.031	0.030	0.029	0.032		0.028	0.026	0.030	0.025
0.492	0.021	0.023	0.024	0.025	0.021	0.025	0.027	0.028	0.025	0.022
0.529	0.030	0.033	0.033	0.032	0.033	0.033	0.034	0.033	0.031	0.029
0.592	0.034	0.033	0.035	0.036	0.034	0.031	0.032	0.033	0.032	0.028
0.655	0.070	0.164	0.054	0.021	0.018	0.069	0.182	0.055	0.027	0.020
0.661	-0.056	-0.257	-0.053	-0.096	-0.099	-0.055		-0.044	-0.095	-0.093
0.687	-0.082	0.014	-0.108	-0.074	-0.051	-0.081	0.016	-0.111		-0.046

(d) α = 3.9 °

0.231	1.188	1.176	1.146	1.095	1.071	0.956	0.952	0.985	1.028	1.080
0.234	0.884	0.866	0.843	0.793	0.758	0.637	0.648	0.677	0.724	0.758
0.246	0.673	0.678	0.643	0.607	0.597	0.548	0.542	0.554	0.579	0.610
0.251	-0.117	-0.117	-0.143	-0.173	-0.194	-0.245	-0.260	-0.235	-0.213	-0.190
0.268	0.320	0.333	0.298	0.263	0.226	0.105	0.099	0.140	0.181	0.212
0.286	0.333	0.309	0.293	0.274	0.260	0.234	0.236	0.252	0.249	0.255
0.292	-0.298	-0.298	-0.306	-0.315	-0.322	-0.327	-0.328	-0.378	-0.326	-0.320
0.322	-0.117	0.072	0.018	-0.160	-0.164	-0.134	-0.168	-0.166	-0.034	0.135
0.352	-0.100	-0.198	-0.171	-0.093	-0.103	-0.116	-0.108	-0.102	-0.171	-0.285
0.381	-0.009	-0.024		-0.017	-0.040	0.068	0.074	0.044	-0.041	-0.026
0.392	0.177	0.154	0.168	0.155	0.158	0.158	0.160	0.159	0.167	0.123
0.402	0.016	0.005	-0.027	0.007	0.008	0.031	0.027	0.010	-0.029	0.023
0.447	0.049	0.043	0.035	0.023	0.009		0.019	0.014	0.024	0.014
0.492	0.040	0.037	0.029	0.021	0.011	0.015	0.015	0.014	0.014	0.011
0.529	0.046	0.046	0.038	0.026	0.023	0.022	0.023	0.021	0.022	0.022
0.592	0.045	0.043	0.040	0.034	0.028	0.024	0.024	0.025	0.025	0.022
0.655	0.093	0.201	0.064	0.022	0.013	0.048	0.122	0.044	0.020	0.014
0.661	-0.040	-0.257	-0.054	-0.098	-0.103	-0.069		-0.054	-0.100	-0.094
0.687	-0.088	0.010	-0.107	-0.079	-0.056	-0.080	0.010	-0.114		-0.050

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TABLE XIII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.28$ - Continued

(e) $\alpha = 6.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	1.238	1.221	1.177	1.100	1.050	0.888	0.886	0.929	0.998	1.069
0.234	0.939	0.916	0.876	0.797	0.742	0.577	0.585	0.626	0.693	0.744
0.246	0.705	0.695	0.657	0.610	0.587	0.519	0.513	0.533	0.560	0.597
0.251	-0.075	-0.073	-0.113	-0.164	-0.198	-0.265	-0.275	-0.252	-0.223	-0.192
0.268	0.372	0.380	0.333	0.287	0.224	0.055	0.041	0.115	0.149	0.202
0.286	0.406	0.382	0.331	0.259	0.212	0.181	0.177	0.168	0.170	0.203
0.292	-0.273	-0.279	-0.296	-0.326	-0.344	-0.343	-0.348	-0.410	-0.360	-0.343
0.322	-0.050	0.187	0.020	-0.174	-0.181	-0.150	-0.197	-0.206	-0.138	0.093
0.352	-0.070	-0.206	-0.142	-0.151	-0.105	-0.131	-0.140	-0.128	-0.195	-0.326
0.381	-0.016	-0.016		-0.102	-0.085	0.083	0.084	0.046	-0.059	-0.031
0.392	0.181	0.159	0.165	0.132	0.135	0.152	0.148	0.145	0.151	0.101
0.402	0.023	0.005	-0.014	-0.024	-0.014	0.030	0.027	-0.001	-0.001	-0.005
0.447	0.062	0.052	0.027	0.004	-0.005		0.009	0.001	-0.006	-0.015
0.492	0.064	0.053	0.027	-0.003	-0.019	0.009	0.002	-0.005	-0.020	-0.022
0.529	0.077	0.070	0.037	-0.003	-0.014	0.014	0.009	-0.001	-0.017	-0.016
0.592	0.081	0.070	0.039	0.001	-0.013	0.013	0.007	-0.000	-0.018	-0.021
0.655	0.161	0.215	0.076	-0.002	-0.026	0.052	0.090	-0.007	-0.021	-0.026
0.661	0.002	-0.244	-0.050	-0.122	-0.140	-0.079		-0.081	-0.140	-0.134
0.687	-0.091	0.049	-0.090	-0.132	-0.118	-0.070	-0.015	-0.111		-0.106

(f) $\alpha = 8.0°$

0.231	1.285	1.265	1.205	1.104	1.045	0.818	0.816	0.874	0.966	1.055
0.234	0.995	0.967	0.907	0.802	0.724	0.509	0.521	0.570	0.655	0.724
0.246	0.734	0.721	0.664	0.608	0.570	0.489	0.483	0.503	0.536	0.578
0.251	-0.029	-0.031	-0.082	-0.155	-0.202	-0.285	-0.293	-0.271	-0.240	-0.195
0.268	0.418	0.423	0.361	0.290	0.217	0.018	-0.000	0.094	0.136	0.197
0.286	0.431	0.404	0.341	0.249	0.190	0.167	0.160	0.136	0.136	0.179
0.292	-0.264	-0.271	-0.296	-0.160	-0.359	-0.349	-0.356	-0.426	-0.379	-0.357
0.322	-0.019	0.230	0.022	-0.181	-0.203	-0.159	-0.210	-0.226	-0.183	0.037
0.352	-0.051	-0.203	-0.122	-0.181	-0.166	-0.131	-0.141	-0.140	-0.141	-0.252
0.381	0.002	-0.012		-0.118	-0.131	0.098	0.091	0.053	-0.059	-0.041
0.392	0.199	0.170	0.160	0.103	0.105	0.153	0.139	0.132	0.101	0.075
0.402	0.019	0.005	-0.010	-0.050	-0.040	0.030	0.025	-0.028	-0.019	-0.036
0.447	0.072	0.059	0.022	-0.006	-0.018		0.005	-0.004	-0.017	-0.030
0.492	0.073	0.060	0.026	-0.014	-0.032	0.007	-0.002	-0.013	-0.033	-0.037
0.529	0.089	0.078	0.036	-0.017	-0.032	0.012	0.002	-0.007	-0.034	-0.037
0.592	0.097	0.081	0.036	-0.016	-0.037	0.008	-0.003	-0.008	-0.038	-0.044
0.655	0.185	0.221	0.076	-0.019	-0.049	0.068	0.084	-0.022	-0.044	-0.053
0.661	0.036	-0.241	-0.053	-0.136	-0.159	-0.063		-0.089	-0.163	-0.159
0.687	-0.080	0.054	-0.083	-0.131	-0.151	-0.087	-0.011	-0.112		-0.135

[redacted]

TABLE XIII.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.28$ - Concluded

(g) $\alpha = 10.0°$

Axial position x/l	Radial position, θ, deg									
	0	0	45	72	90	180	200	225	252	270
0.231	1.326	1.302	1.227	1.101	1.023	0.740	0.739	0.808	0.921	1.030
0.234	1.046	1.013	0.933	0.798	0.696	0.434	0.446	0.502	0.606	0.691
0.246	0.761	0.746	0.676	0.595	0.535	0.455	0.445	0.466	0.494	0.549
0.251	0.021	-0.010	-0.059	-0.148	-0.213	-0.306	-0.316	-0.294	-0.258	-0.201
0.268	0.466	0.465	0.385	0.288	0.207	0.015	-0.046	0.067	0.121	0.188
0.286	0.442	0.410	0.341	0.238	0.171	0.156	0.148	0.120	0.121	0.168
0.292	-0.263	-0.270	-0.300	-0.345	-0.373	-0.358	-0.365	-0.438	-0.391	-0.370
0.322	-0.029	0.207	0.016	-0.185	-0.231	-0.162	-0.212	-0.226	-0.164	-0.041
0.352	-0.066	-0.210	-0.138	-0.164	-0.172	-0.134	-0.143	-0.132	-0.178	-0.241
0.381	-0.019	-0.015		-0.102	-0.175	0.102	0.082	0.056	-0.080	-0.092
0.392	0.219	0.176	0.137	0.067	0.055	0.152	0.116	0.103	0.031	0.034
0.402	0.026	0.003	-0.016	-0.102	-0.083	0.019	0.008	-0.073	-0.043	-0.078
0.447	0.085	0.065	0.012	-0.041	-0.051		-0.011	-0.024	-0.047	-0.067
0.492	0.088	0.070	0.019	-0.038	-0.060	0.004	-0.015	-0.023	-0.053	-0.061
0.529	0.100	0.084	0.032	-0.033	-0.049	0.007	-0.009	-0.014	-0.048	-0.052
0.592	0.099	0.082	0.032	-0.024	-0.045	0.003	-0.007	-0.011	-0.041	-0.046
0.655	0.174	0.216	0.074	-0.017	-0.046	0.058	0.083	-0.019	-0.037	-0.042
0.661	0.020	-0.246	-0.054	-0.131	-0.150	-0.081		-0.085	-0.150	-0.142
0.687	-0.093	0.045	-0.093	-0.165	-0.120	-0.078	-0.018	-0.117		-0.103

[redacted]

TABLE XIV.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.43$

(a) $\alpha = -4.2°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	0.962	0.961	0.994	1.040	1.094	1.230	1.217	1.191	1.131	1.105
0.234	0.613	0.610	0.645	0.699	0.742	0.869	0.880	0.844	0.788	0.742
0.246	0.621	0.635	0.628	0.644	0.659	0.736	0.725	0.706	0.676	0.669
0.251	-0.124	-0.109	-0.119	-0.101	-0.088	-0.021	-0.030	-0.045	-0.071	-0.084
0.268	0.061	0.083	0.097	0.156	0.181	0.291	0.298	0.272	0.211	0.179
0.286	0.181	0.158	0.168	0.195	0.216	0.306	0.302	0.292	0.242	0.213
0.292	-0.273	-0.271	-0.267	-0.256	-0.247	-0.202	-0.208	-0.251	-0.236	-0.246
0.322	-0.137	-0.007	0.006	-0.153	-0.142	-0.088	-0.099	-0.112	-0.001	0.155
0.352	-0.099	-0.165	-0.178	-0.087	-0.080	-0.084	-0.049	-0.051	-0.170	-0.254
0.381	0.001	-0.038		-0.075	-0.092	-0.029	-0.061	-0.099	-0.047	-0.047
0.392	0.125	0.119	0.108	0.122	0.111	0.142	0.130	0.129	0.114	0.091
0.402	0.015	-0.046	0.017	-0.000	0.005	0.015	0.017	0.015	-0.000	0.013
0.447	0.014	0.016	0.009	0.006	0.008		0.032	0.020	0.014	0.004
0.492	0.014	0.006	0.001	0.003	0.003	0.040	0.039	0.029	0.014	0.004
0.529	0.012	0.015	0.011	0.008	0.012	0.044	0.043	0.031	0.016	0.006
0.592	0.011	0.011	0.008	0.009	0.012	0.049	0.047	0.037	0.019	0.007
0.655	0.046	0.124	0.024	0.007	0.009	0.106	0.211	0.062	0.027	0.013
0.661	-0.042	-0.236	-0.055	-0.091	-0.088	-0.003		-0.020	-0.076	-0.075
0.687	-0.071	0.011	-0.099	-0.062	-0.057	-0.088	0.055	-0.082		-0.050

(b) $\alpha = -2.2°$

0.231	1.031	1.027	1.043	1.064	1.101	1.172	1.163	1.151	1.117	1.111
0.234	0.684	0.680	0.700	0.728	0.751	0.805	0.821	0.805	0.777	0.751
0.246	0.652	0.666	0.655	0.660	0.674	0.707	0.699	0.696	0.682	0.680
0.251	-0.104	-0.090	-0.102	-0.090	-0.084	-0.052	-0.057	-0.062	-0.076	-0.081
0.268	0.109	0.128	0.132	0.171	0.180	0.240	0.243	0.233	0.192	0.176
0.286	0.212	0.189	0.194	0.211	0.223	0.266	0.264	0.266	0.236	0.220
0.292	-0.258	-0.257	-0.253	-0.247	-0.242	-0.220	-0.224	-0.264	-0.239	-0.241
0.322	-0.131	0.015	0.011	-0.140	-0.132	-0.107	-0.115	-0.121	-0.006	0.170
0.352	-0.095	-0.170	-0.165	-0.079	-0.066	-0.094	-0.058	-0.065	-0.158	-0.226
0.381	-0.024	-0.035		-0.060	-0.081	-0.035	-0.067	-0.075	-0.023	-0.045
0.392	0.128	0.122	0.119	0.133	0.132	0.133	0.135	0.137	0.124	0.112
0.402	0.013	-0.032	0.018	0.011	0.016	0.015	0.019	0.019	0.014	0.018
0.447	0.018	0.020	0.016	0.017	0.019		0.025	0.019	0.021	0.012
0.492	0.010	0.011	0.008	0.012	0.011	0.030	0.030	0.026	0.020	0.011
0.529	0.016	0.020	0.019	0.019	0.021	0.032	0.034	0.029	0.023	0.017
0.592	0.017	0.016	0.016	0.020	0.022	0.037	0.037	0.034	0.025	0.018
0.655	0.054	0.151	0.038	0.018	0.018	0.089	0.197	0.059	0.034	0.023
0.661	-0.040	-0.213	-0.045	-0.082	-0.079	-0.013		-0.023	-0.070	-0.067
0.687	-0.075	0.019	-0.098	-0.052	-0.045	-0.081	0.040	-0.086		-0.040

[redacted]

TABLE XIV.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.4$ - Continued

(c) $\alpha = -0.1°$

Axial position x/l	Radial position, θ, deg									
	0	20	45	72	90	180	200	225	252	270
0.231	1.097	1.088	1.090	1.084	1.102	1.111	1.102	1.105	1.100	1.113
0.234	0.753	0.743	0.750	0.752	0.754	0.744	0.759	0.760	0.762	0.753
0.246	0.684	0.695	0.676	0.674	0.676	0.678	0.671	0.678	0.678	0.685
0.251	-0.078	-0.071	-0.083	-0.081	-0.082	-0.080	-0.086	-0.081	-0.082	-0.079
0.268	0.160	0.178	0.171	0.188	0.182	0.188	0.188	0.193	0.181	0.176
0.286	0.244	0.220	0.218	0.224	0.227	0.205	0.239	0.249	0.231	0.223
0.292	-0.243	-0.245	-0.243	-0.241	-0.240	-0.232	-0.234	-0.271	-0.240	-0.239
0.322	-0.126	0.025	0.013	-0.136	-0.130	-0.112	-0.120	-0.123	-0.008	0.169
0.352	-0.094	-0.171	-0.164	-0.079	-0.067	-0.093	-0.057	-0.064	-0.158	-0.226
0.381	-0.017	-0.035		-0.064	-0.076	-0.031	-0.062	-0.056	-0.016	-0.041
0.392	0.130	0.124	0.129	0.134	0.139	0.129	0.138	0.136	0.129	0.121
0.402	0.010	-0.015	-0.000	0.016	0.022	0.017	0.019	0.019	0.021	0.016
0.447	0.023	0.024	0.023	0.022	0.023		0.019	0.017	0.021	0.018
0.492	0.018	0.018	0.017	0.019	0.015	0.022	0.023	0.023	0.020	0.018
0.529	0.024	0.027	0.027	0.025	0.026	0.024	0.026	0.026	0.025	0.023
0.592	0.025	0.023	0.025	0.026	0.025	0.027	0.028	0.029	0.026	0.021
0.655	0.069	0.160	0.048	0.023	0.021	0.071	0.173	0.052	0.032	0.026
0.661	-0.031	-0.199	-0.040	-0.079	-0.077	-0.029		-0.027	-0.071	-0.066
0.687	-0.081	0.022	-0.097	-0.051	-0.042	-0.078	0.029	-0.089		-0.037

(d) $\alpha = 1.9°$

0.231	1.162	1.152	1.137	1.108	1.103	1.050	1.044	1.061	1.083	1.115
0.234	0.822	0.809	0.800	0.775	0.754	0.672	0.695	0.711	0.739	0.755
0.246	0.716	0.725	0.698	0.682	0.675	0.652	0.643	0.655	0.669	0.687
0.251	-0.049	-0.043	-0.061	-0.072	-0.082	-0.101	-0.103	-0.097	-0.088	-0.077
0.268	0.216	0.230	0.212	0.207	0.185	0.143	0.140	0.159	0.160	0.176
0.286	0.286	0.262	0.248	0.235	0.224	0.194	0.195	0.214	0.215	0.223
0.292	-0.221	-0.222	-0.227	-0.234	-0.241	-0.254	-0.255	-0.291	-0.247	-0.238
0.322	-0.102	0.085	0.018	-0.128	-0.132	-0.132	-0.144	-0.142	-0.046	0.148
0.352	-0.080	-0.177	-0.149	-0.091	-0.071	-0.101	-0.082	-0.092	-0.166	-0.248
0.381	-0.032	-0.025		-0.057	-0.075	-0.014	-0.036	-0.036	-0.026	-0.046
0.392	0.136	0.130	0.146	0.130	0.136	0.129	0.138	0.133	0.140	0.116
0.402	0.181	-0.000	-0.013	0.016	0.020	-0.000	0.020	0.021	0.003	0.018
0.447	0.032	0.030	0.028	0.023	0.021		0.015	0.012	0.022	0.015
0.492	0.027	0.025	0.021	0.019	0.013	0.015	0.016	0.016	0.019	0.014
0.529	0.034	0.035	0.033	0.026	0.025	0.019	0.021	0.021	0.023	0.020
0.592	0.036	0.034	0.031	0.027	0.023	0.020	0.021	0.021	0.021	0.018
0.655	0.094	0.181	0.058	0.025	0.016	0.050	0.142	0.047	0.025	0.022
0.661	-0.002	-0.190	-0.031	-0.078	-0.082	-0.047		-0.043	-0.077	-0.069
0.687	-0.085	0.048	-0.084	-0.058	-0.048	-0.061	0.005	-0.097		-0.049

[redacted]

TABLE XIV.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.43$ - Continued

(e) $\alpha = 3.9°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	1.220	1.207	1.175	1.122	1.097	0.983	0.977	1.010	1.057	1.108
0.234	0.885	0.867	0.842	0.785	0.746	0.600	0.624	0.656	0.706	0.747
0.246	0.744	0.751	0.716	0.680	0.667	0.619	0.614	0.627	0.650	0.680
0.251	-0.019	-0.016	-0.043	-0.068	-0.086	-0.123	-0.125	-0.116	-0.100	-0.082
0.268	0.272	0.284	0.251	0.225	0.189	0.096	0.091	0.123	0.146	0.178
0.286	0.326	0.299	0.275	0.243	0.220	0.162	0.164	0.188	0.200	0.217
0.292	-0.203	-0.206	-0.216	-0.232	-0.245	-0.268	-0.268	-0.305	-0.255	-0.242
0.322	-0.086	0.109	0.020	-0.130	-0.138	-0.134	-0.151	-0.151	-0.061	0.131
0.352	-0.074	-0.178	-0.145	-0.100	-0.079	-0.103	-0.090	-0.099	-0.178	-0.271
0.381	-0.030	-0.024		-0.061	-0.083	0.007	0.003	-0.024	-0.036	-0.052
0.392	0.144	0.134	0.152	0.118	0.119	0.126	0.132	0.126	0.138	0.099
0.402	0.012	0.010	-0.014	0.007	0.010	0.022	0.019	0.018	-0.012	0.019
0.447	0.039	0.035	0.028	0.017	0.002		0.013	0.007	0.016	0.006
0.492	0.038	0.034	0.024	0.014	0.005	0.021	0.011	0.011	0.006	0.007
0.529	0.045	0.044	0.034	0.020	0.017	0.015	0.015	0.013	0.002	0.012
0.592	0.048	0.043	0.034	0.023	0.015	0.016	0.016	0.015	0.012	0.003
0.655	0.109	0.192	0.062	0.021	0.003	0.043	0.128	0.043	0.019	0.018
0.661	0.008	-0.187	-0.029	-0.081	-0.087	-0.053		-0.048	-0.083	-0.074
0.687	-0.084	0.053	-0.081	-0.064	-0.053	-0.061	-0.001	-0.101		-0.058

(f) $\alpha = 6.0°$

0.231	1.273	1.255	1.209	1.127	1.085	0.908	0.905	0.950	1.024	1.094
0.234	0.945	0.923	0.876	0.789	0.727	0.522	0.546	0.593	0.667	0.728
0.246	0.770	0.762	0.726	0.677	0.651	0.584	0.575	0.598	0.624	0.662
0.251	0.011	0.011	-0.026	-0.067	-0.095	-0.146	-0.148	-0.139	-0.116	-0.092
0.268	0.327	0.334	0.289	0.241	0.190	0.049	0.040	0.087	0.130	0.179
0.286	0.367	0.338	0.299	0.245	0.207	0.128	0.130	0.152	0.172	0.203
0.292	-0.185	-0.190	-0.207	-0.235	-0.255	-0.285	-0.288	-0.277	-0.273	-0.254
0.322	-0.054	0.167	0.022	-0.142	-0.165	-0.138	-0.166	-0.177	-0.118	0.097
0.352	-0.050	-0.179	-0.129	-0.130	-0.100	-0.111	-0.110	-0.119	-0.214	-0.294
0.381	-0.008	-0.018		-0.083	-0.102	0.030	0.028	-0.020	-0.054	-0.062
0.392	0.159	0.137	0.149	0.101	0.087	0.122	0.120	0.117	0.125	0.076
0.402	0.015	0.013	-0.006	-0.021	-0.011	0.022	0.018	0.002	-0.049	-0.005
0.447	0.048	0.039	0.019	0.001	-0.005		0.008	-0.001	-0.007	-0.013
0.492	0.049	0.041	0.021	0.001	-0.011	0.006	0.003	-0.000	-0.010	-0.011
0.529	0.057	0.052	0.032	0.005	-0.001	0.010	0.007	-0.000	-0.008	-0.009
0.592	0.063	0.054	0.032	0.006	-0.007	0.009	0.005	-0.001	-0.012	-0.014
0.655	0.144	0.210	0.069	0.005	-0.014	0.050	0.108	0.010	-0.007	-0.010
0.661	0.042	-0.178	-0.025	-0.096	-0.109	-0.046		-0.052	-0.110	-0.102
0.687	-0.078	0.057	-0.083	-0.097	-0.085	-0.072	-0.006	-0.101		-0.093

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TABLE XIV.- PRESSURE COEFFICIENTS OVER THE BODY WITH NOSE N_7 FOR $M_\infty = 1.43$ - Concluded

(g) $\alpha = 8.0°$

Axial position x/l	Radial position, θ, deg									
	0	30	45	72	90	180	200	225	252	270
0.231	1.324	1.303	1.240	1.134	1.072	0.839	0.837	0.896	0.993	1.082
0.234	1.007	0.978	0.913	0.794	0.707	0.439	0.462	0.523	0.622	0.709
0.246	0.801	0.787	0.731	0.671	0.631	0.551	0.542	0.564	0.597	0.637
0.251	0.045	0.044	-0.000	-0.061	-0.103	-0.161	-0.165	-0.154	-0.130	-0.098
0.268	0.385	0.388	0.328	0.258	0.193	0.020	0.005	0.061	0.124	0.182
0.286	0.414	0.384	0.327	0.248	0.194	0.108	0.103	0.119	0.147	0.189
0.292	-0.165	-0.172	-0.196	-0.235	-0.263	-0.292	-0.299	-0.347	-0.286	-0.260
0.322	-0.023	0.212	0.028	-0.149	-0.185	-0.138	-0.173	-0.197	-0.160	0.049
0.352	-0.027	-0.174	-0.108	-0.149	-0.138	-0.114	-0.120	-0.131	-0.220	-0.270
0.381	0.013	-0.007		-0.116	-0.120	0.047	0.045	-0.011	-0.079	-0.079
0.392	0.186	0.149	0.137	0.090	0.050	0.125	0.113	0.107	0.062	0.043
0.402	0.034	0.017	0.006	-0.046	-0.043	0.024	0.019	-0.025	-0.019	-0.041
0.447	0.064	0.047	0.003	-0.017	-0.027		0.003	-0.008	-0.023	-0.036
0.492	0.065	0.053	0.021	-0.015	-0.031	0.005	-0.003	-0.013	-0.030	-0.033
0.529	0.077	0.067	0.031	-0.013	-0.025	0.009	-0.000	-0.008	-0.028	-0.031
0.592	0.084	0.070	0.032	-0.011	-0.031	0.007	-0.005	-0.008	-0.033	-0.038
0.655	0.170	0.220	0.071	-0.010	-0.036	0.064	0.096	-0.005	-0.030	-0.036
0.661	0.063	-0.174	-0.027	-0.111	-0.129	-0.031		-0.061	-0.132	-0.127
0.687	-0.066	0.059	-0.083	-0.125	-0.116	-0.130	0.002	-0.101		-0.123

(h) $\alpha = 10.0°$

0.231	1.371	1.347	1.267	1.137	1.056	0.777	0.776	0.842	0.957	1.066
0.234	1.065	1.031	0.946	0.797	0.681	0.341	0.339	0.431	0.569	0.680
0.246	0.832	0.815	0.747	0.661	0.598	0.518	0.507	0.529	0.557	0.615
0.251	0.092	0.088	0.019	-0.055	-0.109	-0.171	-0.181	-0.169	-0.145	-0.105
0.268	0.438	0.438	0.362	0.271	0.192	-0.016	-0.033	0.040	0.114	0.177
0.286	0.458	0.423	0.221	0.249	0.179	0.092	0.081	0.091	0.124	0.175
0.292	-0.145	-0.154	-0.186	-0.235	-0.269	-0.298	-0.306	-0.362	-0.297	-0.268
0.322	-0.000	0.238	0.035	-0.150	-0.196	-0.136	-0.175	-0.209	-0.180	-0.006
0.352	-0.012	-0.167	-0.094	-0.156	-0.189	-0.111	-0.120	-0.134	-0.216	-0.192
0.381	0.020	-0.000		-0.130	-0.142	0.053	0.044	-0.000	-0.141	-0.139
0.392	0.216	0.168	0.109	0.063	0.007	0.131	0.098	0.088	0.021	-0.000
0.402	0.059	0.027	-0.010	-0.068	-0.094	0.024	0.010	-0.056	-0.046	-0.073
0.447	0.080	0.059	0.002	-0.048	-0.057		-0.009	-0.026	-0.048	-0.068
0.492	0.084	0.065	0.016	-0.040	-0.059	0.006	-0.014	-0.021	-0.052	-0.060
0.529	0.096	0.080	0.029	-0.035	-0.051	0.008	-0.012	-0.015	-0.047	-0.056
0.592	0.101	0.082	0.031	-0.027	-0.050	0.006	-0.015	-0.015	-0.048	-0.055
0.655	0.180	0.227	0.073	-0.021	-0.050	0.067	0.092	-0.008	-0.038	-0.046
0.661	0.074	-0.168	-0.025	-0.117	-0.137	-0.025		-0.058	-0.135	-0.132
0.687	-0.058	0.063	-0.079	-0.129	-0.120	0.013	0.009	-0.097		-0.125

[redacted]

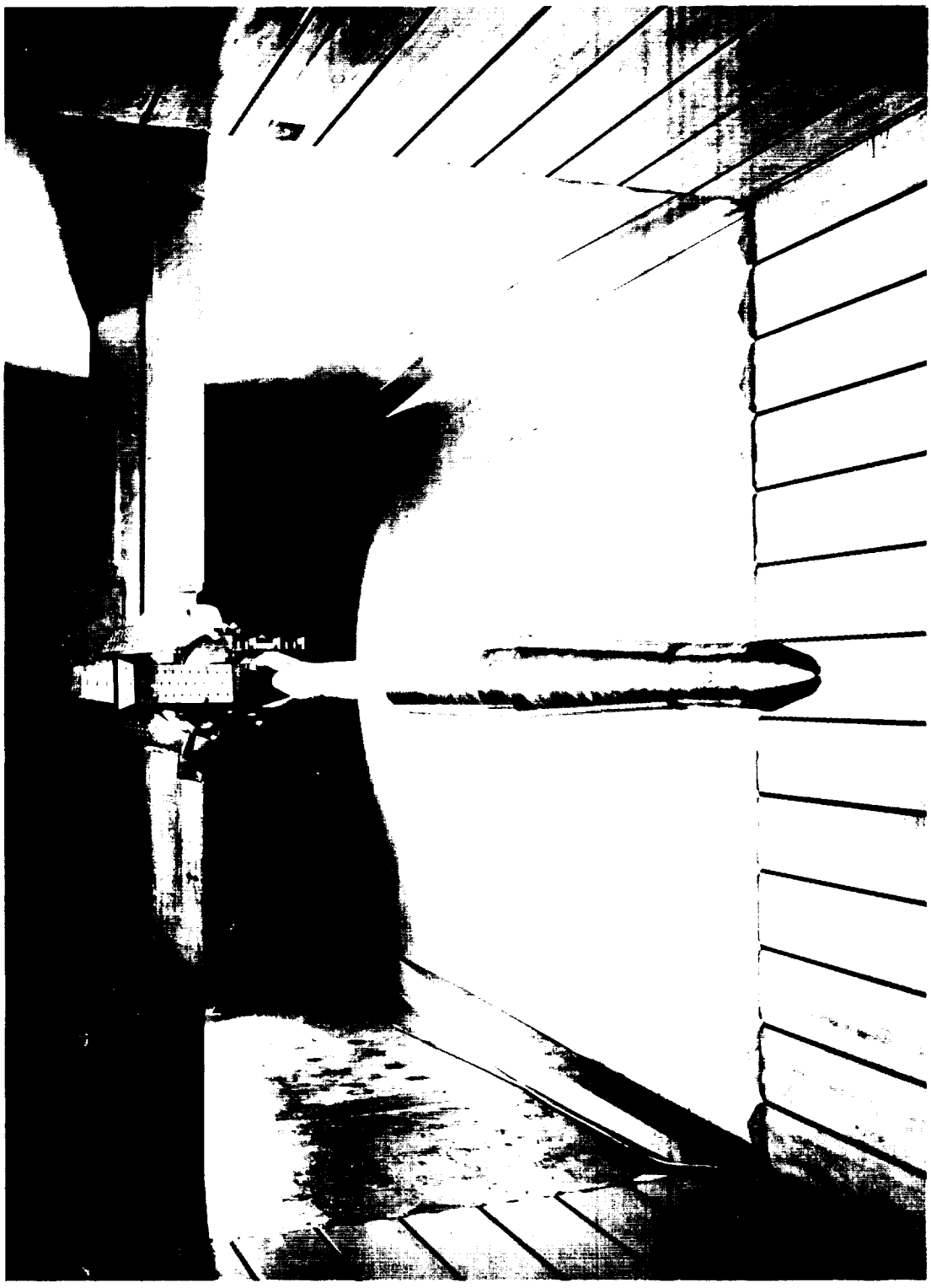

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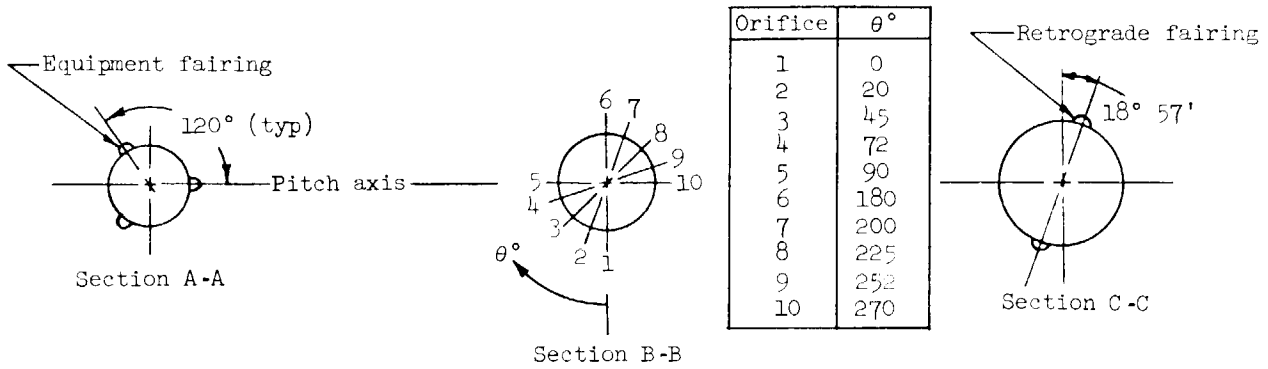
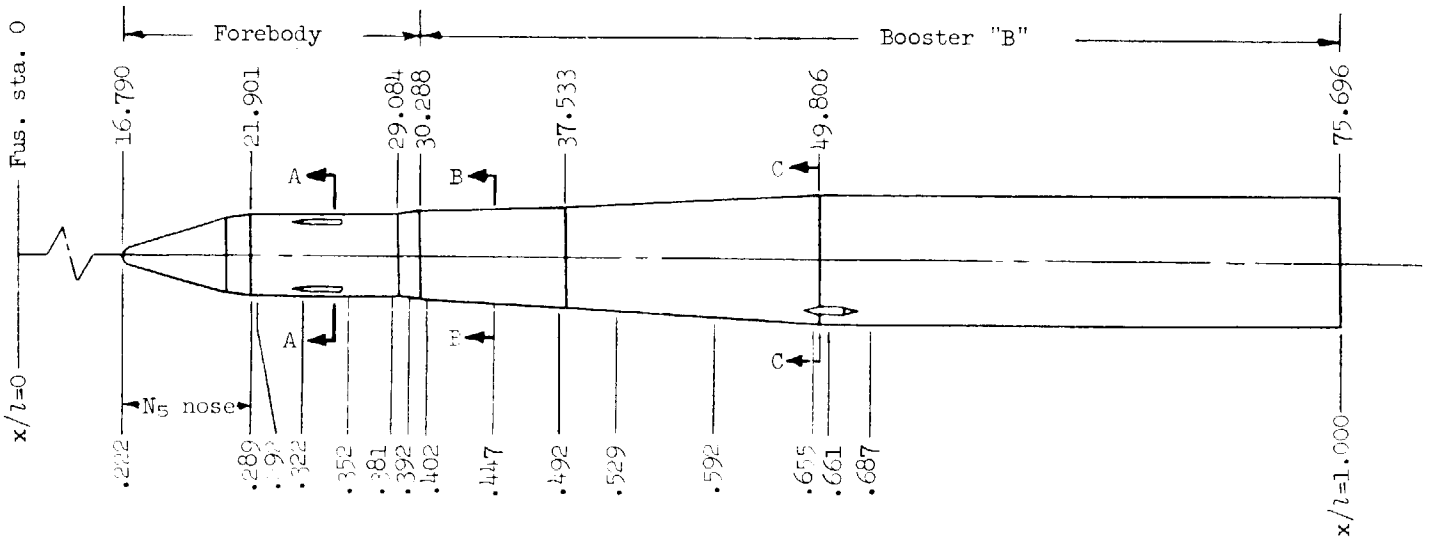
Figure 1.- Photograph of model in the Ames 11- by 11-foot transonic test section.

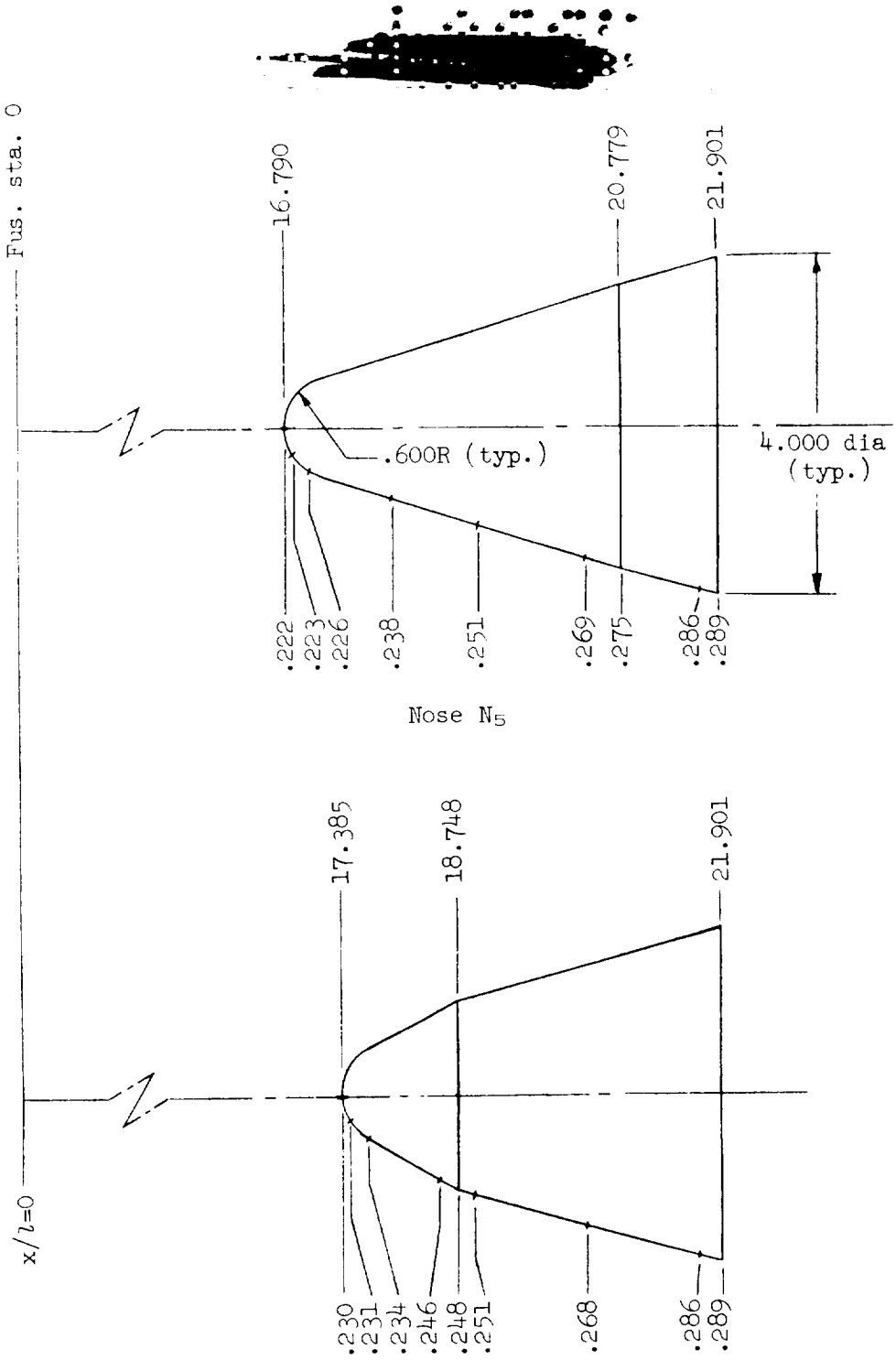


Orifice	θ°
1	0
2	20
3	45
4	72
5	90
6	180
7	200
8	225
9	252
10	270



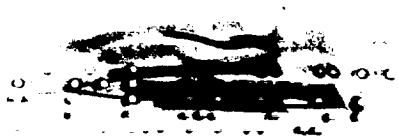
Note: Fuselage stations in inches.

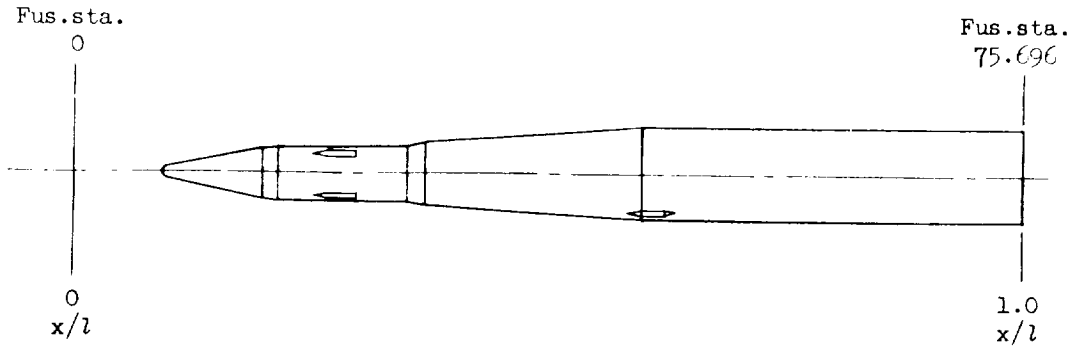
Figure 2.- General arrangement of the model.



Note: Fuselage stations in inches.

Figure 3.- Nose details of the model.





N_5 nose				N_7 nose			
x/l	Fuselage station	Surface angle, deg	Local body dia, in.	x/l	Fuselage station	Surface angle, deg	Local body dia, in.
.2[illegible]	16.790	90.0	0	.230	17.385	90.0	0
.2[illegible]3	16.864	60.0	0.600	.231	17.472	60.0	0.600
.2[illegible]6	17.074	30.0	1.040	.234	17.706	31.3	1.010
.238	18.031	17.5	1.659	.246	18.593	31.3	[illegible].088
.2[illegible]1	19.001	17.5	2.267	.251	18.972	15.0	2.434
.[illegible]60	[illegible]0.341	17.5	3.114	.268	[illegible]0.311	15.0	3.154
.286	[illegible]1.683	15.0	3.870	.286	21.654	15.0	3.870
.2[illegible]2	[illegible]2.126	0	4.000	.292	22.093	0	4.000
.3[illegible]	24.359	0	4.000	Same as N_5			
.35[illegible]	26.612	0	4.000				
.381	28.821	0	4.000				
.3[illegible]	29.679	5.8	4.122				
.40[illegible]	30.457	3.0	4.265				
.447	33.844	3.0	4.518				
.4[illegible]	37.[illegible]62	3.0	4.[illegible]74				
.52[illegible]	40.012	3.2	5.[illegible]84				
.59[illegible]	44.792	3.2	5.8[illegible]8				
.655	49.585	3.2	6.376				
.661	50.025	0	6.400				
.687	52.031	0	6.400				

Figure 4.- Pressure orifice locations on the model.

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