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MEMORANDUM

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THE VERTICAL-TAIL PRESSURE DISTRIBUTIONS OF A COMPLETE

MODEL IN SIDESLIP AT HIGH SUBSONIC SPEEDS

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SUMMARY

An investigation has been made in the Langley high-speed 7- by 10-foot tunnel of some effects of horizontal-tail position on the vertical-tail pressure distributions of a complete model in sideslip at high subsonic speeds. The wing of the model was swept back 28.82° at the quarter-chord line and had an aspect ratio of 3.50, a taper ratio of 0.067, and NACA 65A004 airfoil sections parallel to the model plane of symmetry. Tests were made with the horizontal tail off, on the wing-chord plane extended, and in T-tail arrangements in forward and rearward locations. The test Mach numbers ranged from 0.60 to 0.92, which corresponds to a Reynolds number range from approximately 2.93×10^6 to 3.69×10^6, based on the wing mean aerodynamic chord. The sideslip angles varied from -3.9° to 12.7° at several selected angles of attack.

The results indicated that, for a given angle of sideslip, increases in angle of attack caused reductions in the vertical-tail loads in the vicinity of the root chord and increases at the midspan and tip locations, with rearward movements in the local chordwise centers of pressure for the midspan locations and forward movements near the tip of the vertical tail. At the higher angles of attack all configurations investigated experienced outboard and rearward shifts in the center of pressure of the total vertical-tail load. Location of the horizontal tail on the wing-chord plane extended produced only small effects on the vertical-tail loads and centers of pressure. Locating the horizontal tail at the tip of the vertical tail in the forward position caused increases in the vertical-tail loads; this configuration, however, experienced considerable reduction in loads with increasing Mach number. Location of the horizontal tail at the tip of the vertical tail in the rearward position produced the largest increases in vertical-tail loads per degree sideslip angle; this configuration experienced the smallest variations of loads with Mach number of any of the configurations investigated.

INTRODUCTION

The need for a better understanding of the nature of the air flow at the tail and its effects on the resulting vertical-tail loads has become increasingly important because several high-speed airplanes in current use have, in maneuvering flight, been subjected to extreme vertical-tail loads and experienced losses in directional stability. Of the many variables affecting the aerodynamic characteristics of a particular vertical-tail configuration, wing position and fuselage cross-sectional shape have been studied and the results of these investigations have been presented in references 1 to 5. A recent experimental investigation (ref. 6) has shown that the directional stability of a complete airplane model is also affected appreciably by the location of the horizontal tail.

The purpose of this paper is to present the results of an experimental investigation made to determine the effects of horizontal-tail position on the chordwise pressure distributions and integrated loads on the vertical tail of a complete model in sideslip at high subsonic speeds. The complete model consisted of a highly tapered wing (taper ratio, 0.067) of aspect ratio 3.50 with a quarter-chord sweepback of 28.82°, located on the center line of a fuselage which had an ogival nose, a cylindrical center section, and a truncated tail cone. The vertical-tail aspect ratio was 1.02 and the quarter-chord sweepback angle was 28.35°. The triangular horizontal tail was of aspect ratio 4.00 and was located either on the wing-chord plane or at the tip of the vertical tail in T-tail arrangements. Two chordwise locations were investigated for the T-tail configurations: one with the horizontal tail forward (zero trailing-edge overhang) and one with the horizontal tail rearward (zero leading-edge overhang). The sideslip-angle range extended from -3.9° to 12.7° at angles of attack of approximately -9°, 0°, 9°, and 15°. The test Mach numbers varied from 0.60 to 0.92 with the corresponding Reynolds number (based on wing mean aerodynamic chord) varying from 2.93×10^6 to 3.69×10^6.

SYMBOLS

C_L lift coefficient, $\dfrac{\text{Lift}}{qS}$

S wing area, sq ft

C_p pressure coefficient, $\dfrac{p_l - p_\infty}{q}$

p_l local static pressure, lb/sq ft

p_∞ free-stream static pressure, lb/sq ft

q dynamic pressure, $\frac{1}{2}\rho V_\infty^2$, lb/sq ft

ρ air density, slugs/cu ft

V_∞ free-stream velocity, ft/sec

M Mach number

c_v local vertical-tail chord, ft

$\bar{c}_v$ vertical-tail mean aerodynamic chord, ft

$c_{v,av}$ average vertical-tail chord, ft

$\bar{c}$ wing mean aerodynamic chord, ft

b_v exposed vertical-tail span, ft

x chordwise distance from leading edge of vertical-tail local chord, ft

z spanwise distance from vertical-tail root chord (vertical-tail root chord 0.154 ft above fuselage center line), ft

z' spanwise distance from vertical-tail root chord to vertical-tail mean aerodynamic chord, ft

$\Delta\left(\dfrac{x}{c_v}\right)$ increment of vertical-tail local chord over which the pressure at a particular orifice is assumed to act (distance between points midway between adjacent orifices)

$c_{n,v}$ vertical-tail section normal-force coefficient,

$$\sum_{\frac{x}{c_v}=0}^{1} \left(C_{p,L} - C_{p,R}\right)\Delta\left(\frac{x}{c_v}\right)$$

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$c_{m,v}$ vertical-tail section pitching-moment coefficient about 0.25 local vertical-tail chord,

$$\sum_{\frac{x}{c_v}=0}^{1} \left(C_{p,L} - C_{p,R}\right)\left(\frac{x}{c_v} - 0.25\right)\Delta\left(\frac{x}{c_v}\right)$$

$\dfrac{\bar{x}}{c_v}$ local chordwise location of vertical-tail center of pressure,

$$-\left(\frac{c_{m,v}}{c_{n,v}} - 0.25\right)$$

$\dfrac{\bar{x}}{\bar{c}_v}$ chordwise center-of-pressure location of total vertical-tail load,

$$0.25 - \frac{\displaystyle\int_0^{1.0} \frac{\Delta c_n c_v}{\Delta\beta c_{v,av}}\left[\left(\frac{z'}{b_v} - \frac{z}{b_v}\right)\frac{b_v}{\bar{c}_v}\tan\Lambda + \left(0.25 - \frac{\bar{x}}{c_v}\right)\frac{c_v}{\bar{c}_v}\right]d\left(\frac{z}{b_v}\right)}{\displaystyle\int_0^{1.0} \frac{\Delta c_n c_v}{\Delta\beta c_{v,av}}\, d\left(\frac{z}{b_v}\right)}$$

$\dfrac{\bar{z}}{b_v}$ spanwise center-of-pressure location of total vertical-tail load,

$$\frac{\displaystyle\int_0^{1.0} \frac{\Delta c_n c_v}{\Delta\beta c_{v,av}}\frac{z}{b_v}\, d\left(\frac{z}{b_v}\right)}{\displaystyle\int_0^{1.0} \frac{\Delta c_n c_v}{\Delta\beta c_{v,av}}\, d\left(\frac{z}{b_v}\right)}$$

α angle of attack, deg

β angle of sideslip, deg

i_t horizontal-tail incidence angle, deg

Λ sweep angle of quarter-chord line, deg

Subscripts:

R   right side of vertical tail looking upstream

L   left side of vertical tail looking upstream

MODELS AND APPARATUS

Details of the complete model as tested are presented in figure 1 and a photograph of a typical model mounted on the sting-support system is presented as figure 2. The fuselage had a fineness ratio of 10.94 and was constructed of aluminum. The physical characteristics of the fuselage including afterbody ordinates are given in figure 1(d). The aluminum wing was swept back 28.82° at the quarter-chord line (wing 80-percent-chord line was unswept) and had an aspect ratio of 3.50, a taper ratio of 0.067, and NACA 65A004 airfoil sections parallel to the fuselage center line. The triangular horizontal tail was made of steel and covered with plastic and fiber glass; it had an aspect ratio of 4.00 and NACA 65A006 airfoil sections parallel to the fuselage center line. The horizontal tail was tested in three locations: on the wing chord line, and atop the vertical tail in both forward and rearward locations. These locations are shown in figure 1(b). The vertical tail was made of steel and was covered with plastic and fiber glass; it had an aspect ratio of 1.02 (based on exposed area and span), a taper ratio of 0.46, quarter-chord sweep of 28.35°, and NACA 65A006 airfoil sections. Chordwise pressure orifices (in rows parallel to the fuselage center line) were located at four spanwise stations. Details of the orifice locations are given in figure 1(c) and other details of model geometry are given in table I. The model of this investigation is identical to the model of reference 6.

Tests were made on the sting-support system shown in figure 2. With this system the model can be remotely operated through an angle-of-attack range of approximately 26° in the plane of the vertical strut. By utilization of couplings in the sting behind the model, the model can be rolled 90° so that either angle of attack or angle of sideslip can be the remotely controlled variable. With the wings vertical, the couplings can be used to support the model at fixed angles of attack while the model is tested through the angle-of-sideslip range.

With the models at a given angle of attack and angle of sideslip, a record was taken of the pressures existing at the orifices by photographing a manometer board to which the orifices were connected. These pressures were reduced to coefficient form and are presented in tables II to VIII.

TESTS AND CORRECTIONS

The tests were made in the Langley high-speed 7- by 10-foot tunnel through a Mach number range from 0.60 to 0.92, which corresponds to a Reynolds number range of approximately 2.93×10^6 to 3.69×10^6 based

on the wing mean aerodynamic chord. Pressure measurements on the verti-
cal tail were made through a sideslip-angle range from -3.9° to 12.7° at
angles of attack of approximately -9°, 0°, 9°, and 15° with the horizontal
tail on the fuselage center line and at the tip of the vertical tail in
T-tail arrangements. Two chordwise locations were investigated for the
T-tail configurations: one with the horizontal tail forward (zero trailing-
edge overhang) and one with the horizontal tail rearward (zero leading-edge
overhang). (See fig. 1(b).) Pressure measurements were also made for con-
figurations for which the wing, the horizontal tail, or both, were removed.
The effects of changing the horizontal-tail incidence angle from 0° to -6°
were investigated for the T-tail configuration in the forward position.

Blockage corrections calculated by the method of reference 7 were
applied to Mach number and dynamic pressure. Jet-boundary corrections
calculated by the method of reference 8 have been applied to the angle
of attack. Corrections for sting-support deflection have been applied
to the angles of attack and sideslip.

<div align="center">PRESENTATION OF RESULTS</div>

All the individual pressure coefficients are presented in tables II
to VIII and some selected results from the investigation are presented in
the following figures:

DISCUSSION

Vertical-Tail Loads

For a given angle of attack, increasing the angle of sideslip caused increases in the vertical-tail section loads for all configurations investigated, except at the highest sideslip angle (12.7°) where in some cases load reductions occurred near the tip of the vertical tail (fig. 6).

For all configurations investigated (figs. 7 to 9) the effects of increasing the angle of attack for small angles of sideslip was to cause decreases in the vertical-tail loads in the vicinity of the root chord and to cause increases near the midspan locations. The loss in vertical-tail loads near the root and gain in the midspan region can be attributed to the large flow angularities generated by the fuselage vortices which become stronger and asymmetrically located with respect to the tail plane as the angle of attack is increased (refs. 2 and 9). The addition of the wing on the fuselage center line produced no significant effects on the vertical-tail loads except at the highest angle of attack (approx. 15°, figs. 7 and 8) where the wing causes an outboard shift of the spanwise center of load. This shift in load is presumed to be due to the wing downwash causing the body separation vortices to cross the vertical tail at a lower level. This effect has also been noted in reference 2.

The addition of the horizontal tail on the wing-chord plane extended (figs. 6 to 8) produced only small increases in the vertical-tail loads because of its small end-plate effect. Locating the horizontal tail at the tip of the vertical tail in the forward position (zero trailing-edge overhang, figs. 6 to 9) caused large increases in the vertical-tail loads, particularly near the tip of the vertical tail when compared with those for the horizontal-tail-off configuration or the configuration having the horizontal tail on the wing-chord plane extended. These increases in vertical-tail loads can be attributed to the more complete end-plate effectiveness of the horizontal tail on the vertical tail than to that of the horizontal-tail configuration located on the wing-chord plane extended. Moving the horizontal tail rearward on the vertical tail (zero leading-edge overhang, figs. 6 and 9) produced the largest vertical-tail loads of any of the configurations investigated; this indicated that the horizontal-tail configuration located rearward on the tip of the vertical tail produced a more favorable interference flow field. These results correlate well with the directional stability characteristics presented for the same configurations in reference 6. It should also be pointed out that the effects of the locations of the tip-mounted horizontal-tail configurations on the directional stability (ref. 6) are greater than the effects on the vertical-tail loads and, therefore, the actual loads

experienced by the vertical tail per unit restoring moment are less than for the horizontal tail off or horizontal tail located on wing-chord plane extended.

In general, the effects of change in Mach number on the vertical-tail loads for horizontal tail off and horizontal tail located on the wing-chord plane extended are small (figs. 6 to 8). For the T-tail configuration with the horizontal tail in the forward position, however, increasing the Mach number causes rather large reductions in the vertical-tail loads, particularly at zero angle of attack. The tail-load reductions are somewhat smaller at the higher angles of attack (figs. 7 and 8). The attendant reduction in directional stability for this configuration is noted in reference 6. The reductions in vertical-tail loads are presumed to be due to shock-interference effects between the horizontal and vertical tails. Reducing the horizontal-tail incidence for the horizontal-tail-forward T-tail configuration caused decreases in the vertical-tail loads, with these decreases becoming larger as the Mach number was increased. With the horizontal tail in the rearward position (figs. 6 and 9), the effects of increasing Mach number were to cause only small reductions in the vertical-tail loads, except at the highest Mach number where the loads again increase. These Mach number effects for the horizontal-tail-rearward T-tail configuration are presumed to be due to the shift in the position of the peak minimum pressures associated with the maximum thicknesses of the horizontal tail relative to the vertical-tail lifting pressures and, thereby, the severity of the shock interactions was reduced and more desirable load variations with Mach number were produced for this tail position than for the other horizontal-tail locations investigated.

Vertical-Tail Center of Pressure

In general, for the various horizontal-tail locations, the effects of increasing the angle of sideslip were to cause rearward movements in the local chordwise centers of pressure for the midspan locations and forward movements near the tip of the vertical tail (fig. 10). Increasing the angle of attack also caused rearward movements of the chordwise centers of pressure except at the tip of the vertical tail where some small forward movement was in evidence (fig. 10).

At small angles of attack and sideslip the addition of the horizontal tail tended to reduce the spanwise variations of the vertical-tail local chordwise centers of pressure. As the angles of attack and sideslip were increased, however, the effects of horizontal-tail position decreased and the local centers of pressure were essentially the same for all tail positions (fig. 10).

Increasing Mach number generally caused forward shifts in the vertical-tail chordwise centers of pressures for the lower spanwise locations and a rearward shift in the vicinity of the tip of the vertical tail. The largest Mach number effects were exhibited by the T-tail configuration with the horizontal tail mounted in the forward location and the smallest Mach number effects were exhibited by the T-tail arrangement with the horizontal tail mounted in the rearward location (fig. 10).

The center of pressure of the total vertical-tail load for the wing—fuselage—vertical-tail configuration (fig. 11) at a Mach number of 0.80 was located at approximately 45 percent of the vertical-tail span and 25 percent of the tail mean aerodynamic chord and was essentially invariant with increases in angle of attack to an angle of approximately 10° where further increases in angle caused an outboard and rather abrupt rearward movement.

The addition of the horizontal tail on the wing-chord plane extended produced below 10° angle of attack, only small changes in the spanwise location of the center of pressure (fig. 11(a)) although the chordwise center-of-pressure location (fig. 11(b)) moved rearward gradually with moderate increases in angle of attack. Above $\alpha = 10^{\circ}$, the chord-plane-extended horizontal-tail configuration produced the largest outboard and rearward shifts in center-of-pressure location of any of the configurations investigated. Locating the horizontal tail at the tip of the vertical tail in the forward position caused greater outboard and rearward shifts in the center-of-pressure locations than those of the horizontal-tail-off configuration throughout the moderate angle-of-attack range at a Mach number of 0.80. Locating the horizontal tail in the rear position at the tip of the vertical tail produced outboard and rearward shifts in the center-of-pressure locations, when compared with those of the horizontal-tail-off configuration that were essentially constant throughout the angle-of-attack range. The large outboard shifts in spanwise center-of-pressure location above 10° angle of attack were, as mentioned previously, due to the fuselage vortices which cause flow-angularity reductions at the inboard spanwise locations and increases for the outboard locations.

In general, the effects of increasing Mach number on the locations of the vertical-tail center of pressure were small for all configurations, except for the T-tail arrangement with the horizontal tail in the forward position where increasing Mach number caused a considerable inboard movement.

TABLE I

PHYSICAL CHARACTERISTICS OF THE MODEL

Wing:
Span, ft 2.49
Root chord, ft 1.33
Tip chord, ft 0.089
Mean aerodynamic chord, ft 0.89
Area, sq ft 1.77
Aspect ratio 3.50
Taper ratio 0.067
Quarter-chord sweep, deg 28.82
Airfoil section . . . . NACA 65A004

Horizontal tail:
Span, ft 1.162
Root chord, ft 0.581
Tip chord, ft 0
Mean aerodynamic chord, ft 0.388
Area, sq ft 0.337
Aspect ratio 4.00
Quarter-chord sweep, deg 36.85
Airfoil section . . . . NACA 65A006

Vertical tail:
Span (measured from root chord), ft 0.683
Root chord (located 0.154 ft above fuselage center
    line), ft 0.912
Tip chord, ft 0.420
Mean aerodynamic chord, ft 0.696
Area, sq ft 0.454
Aspect ratio 1.02
Taper ratio 0.46
Quarter-chord sweep, deg 28.35
Airfoil section . . . . NACA 65A006

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL

(a) $\alpha = -9.4°$; $M = 0.60$

$\frac{x}{c_v}$	C_p for —					C_p for —				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.832	.558	.557	-.164	-1.054	.829	.576	.577	-.180	-.780
.025	-.100	-.522	-.530	-.982	-2.458	-.071	-.542	-.549	-1.299	-1.155
.075	-.121	-.342	-.340	-.601	-.992	-.127	-.396	-.400	-.840	-1.120
.150	-.162	-.296	-.301	-.441	-.672	-.141	-.305	-.306	-.496	-1.054
.250	-.123	-.203	-.207	-.279	-.396	-.150	-.257	-.257	-.361	-.964
.350	-.123	-.175	-.179	-.217	-.302	-.175	-.241	-.244	-.301	-.792
.450	-.137	-.164	-.166	-.185	-.251	-.189	-.219	-.223	-.244	-.585
.550	-.109	-.127	-.127	-.126	-.192	-.148	-.157	-.161	-.169	-.424
.650	-.068	-.073	-.079	-.071	-.141	-.102	-.105	-.106	-.107	-.309
.750	-.027	-.016	-.019	-.014	-.100	-.039	-.045	-.042	-.043	-.203
.850	.027	.037	.036	.039	-.033	.021	.025	.032	.032	-.097
.900	.048	.066	.066	.064	.008	.059	.062	.064	.062	-.040
Right side										
.025	-.064	.276	.275	.539	.790	-.086	.340	.341	.596	.769
.075	-.150	.082	.082	.285	.498	-.143	.119	.123	.331	.512
.150	-.162	.002	.002	.151	.316	-.150	.018	.025	.183	.237
.250	-.127	-.027	-.028	.080	.204	-.164	-.041	-.040	.080	.199
.350	-.139	-.068	-.067	.011	.105	-.175	-.084	-.088	.009	.103
.450	-.148	-.105	-.104	-.041	.027	-.171	-.118	-.115	-.043	.018
.550	-.112	-.084	-.088	-.046	-.001	-.143	-.107	-.108	-.066	-.017
.650	-.066	-.055	-.053	-.030	-.001	-.102	-.075	-.074	-.037	-.015
.750	-.023	-.016	-.021	.002	.008	-.025	-.025	-.029	-.002	.002
.850	.043	.048	.043	.052	.050	.046	.057	.059	.059	.041
.900	.062	.062	.064	.066	.061	.055	.057	.057	.062	.034
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.829	.545	.550	-.016	-.373	.809	.515	.518	.062	-.173
.025	-.093	-.667	-.670	-1.012	-.769	-.189	-.615	-.620	-.875	-.534
.075	-.139	-.440	-.441	-.938	-.752	-.157	-.373	-.374	-.854	-.552
.150	-.162	-.344	-.349	-.685	-.739	-.148	-.278	-.276	-.617	-.585
.250	-.166	-.273	-.276	-.425	-.700	-.096	-.200	-.200	-.375	-.594
.350	-.180	-.246	-.248	-.322	-.647					
.450	-.182	-.223	-.225	-.256	-.582					
.550	-.159	-.184	-.186	-.201	-.511					
.650	-.121	-.121	-.122	-.133	-.433	-.107	-.153	-.154	-.297	-.431
.750	-.036	-.052	-.051	-.066	-.352	-.059	-.121	-.115	-.345	-.394
.850	.002	.007	.004	-.007	-.279	-.025	-.086	-.081	-.375	-.355
.900	.023	.030	.027	.023	-.238	.023	-.045	-.044	-.370	-.336
Right side										
.025	-.098	.362	.371	.614	.753	-.143	.264	.268	.470	.581
.075	-.159	.123	.126	.342	.503	-.175	.037	.032	.176	.291
.150	-.159	.034	.034	.192	.328	-.137	-.014	-.019	.066	.151
.250	-.180	-.043	-.044	.073	.188					
.350	-.189	-.098	-.097	-.002	.084					
.450	-.187	-.123	-.120	-.048	.011					
.550	-.159	-.118	-.115	-.071	-.021					
.650	-.109	-.086	-.081	-.043	-.031	-.089	-.071	-.072	-.043	-.035
.750	-.039	-.039	-.035	-.012	-.031	-.057	-.050	-.051	-.037	-.072
.850	.021	.041	.045	.052	-.015	.000	.002	.004	.007	-.063
.900	.050	.052	.057	.055	-.031	.014	.014	.011	.004	-.090

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continued

(b) $\alpha = -9.6°$; M = 0.80

$\frac{x}{c_v}$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v=0.11$					$z/b_v=0.38$				
Left side										
.000	.851	.879	.718	.352	-.129	.655	.862	.735	.357	-.147
.025	.238	-.104	-.541	-.891	-1.541	.322	-.075	-.918	-1.840	-1.132
.075	.103	-.128	-.375	-.804	-1.223	.123	-.147	-.430	-1.419	-1.100
.150	-.010	-.188	-.343	-.513	-.964	.022	-.168	-.342	-.402	-.978
.250	-.037	-.147	-.224	-.295	-.573	-.051	-.190	-.287	-.318	-.849
.350	-.067	-.142	-.195	-.223	-.381	-.117	-.207	-.271	-.284	-.683
.450	-.117	-.162	-.181	-.188	-.292	-.172	-.227	-.244	-.243	-.542
.550	-.103	-.128	-.131	-.132	-.223	-.143	-.166	-.174	-.164	-.431
.650	-.074	-.071	-.071	-.065	-.168	-.100	-.115	-.105	-.092	-.344
.750	-.030	-.019	-.010	-.004	-.092	-.039	-.037	-.027	-.019	-.251
.850	.034	.040	.054	.053	-.017	.037	.038	.053	.058	-.153
.900	.062	.075	.091	.085	.019	.076	.079	.094	.093	-.097
Right side										
.025	-.420	-.028	.285	.535	.764	-.894	-.075	.340	.581	.757
.075	-.411	-.148	.083	.282	.480	-.449	-.154	.115	.317	.500
.150	-.346	-.175	-.002	.143	.305	-.355	-.171	.019	.170	.325
.250	-.242	-.138	-.025	.073	.198	-.285	-.182	-.054	.064	.186
.350	-.213	-.156	-.077	-.002	.091	-.261	-.197	-.112	-.017	.082
.450	-.193	-.171	-.125	-.065	-.004	-.233	-.201	-.148	-.083	-.014
.550	-.131	-.125	-.106	-.074	-.037	-.178	-.165	-.134	-.101	-.055
.650	-.063	-.071	-.065	-.048	-.036	-.097	-.107	-.094	-.066	-.051
.750	-.007	-.016	-.019	-.019	-.028	-.013	-.021	-.019	-.014	-.031
.850	.067	.055	.053	.044	.021	.070	.061	.067	.054	.009
.900	.100	.081	.076	.068	.038	.097	.070	.074	.067	-.004
	$z/b_v=0.66$					$z/b_v=0.93$				
Left side										
.000	.637	.870	.721	.346	-.040	.709	.849	.667	.317	.013
.025	.345	-.107	-1.097	-1.710	-.830	.233	-.210	-1.094	-.990	-.576
.075	.143	-.163	-.466	-1.495	-.849	.031	-.198	-.427	-.910	-.591
.150	.016	-.195	-.395	-.828	-.814	-.051	-.172	-.296	-.736	-.614
.250	-.065	-.203	-.305	-.355	-.718	-.024	-.116	-.218	-.488	-.597
.350	-.128	-.216	-.273	-.295	-.663					
.450	-.163	-.221	-.245	-.240	-.588					
.550	-.152	-.185	-.193	-.182	-.497					
.650	-.106	-.141	-.117	-.110	-.408	-.096	-.116	-.158	-.323	-.419
.750	-.051	-.030	-.036	-.040	-.333	-.044	-.051	-.112	-.359	-.378
.850	.022	.019	.028	.018	-.255	-.008	-.006	-.082	-.422	-.336
.900	.047	.049	.063	.047	-.225	.010	.016	-.045	-.407	-.320
Right side										
.025	-1.122	-.087	.371	.601	.760	-1.082	-.156	.261	.462	.598
.075	-.489	-.174	.128	.326	.500	-.438	-.207	.004	.140	.290
.150	-.372	-.180	.025	.177	.329	-.276	-.159	-.039	.027	.138
.250	-.314	-.204	-.067	.048	.178					
.350	-.288	-.218	-.129	-.043	.056					
.450	-.247	-.213	-.161	-.095	-.022					
.550	-.189	-.185	-.148	-.112	-.062					
.650	-.111	-.116	-.100	-.071	-.065	-.114	-.089	-.071	-.062	-.065
.750	-.033	-.033	-.028	-.030	-.057	-.129	-.042	-.044	-.043	-.083
.850	.054	.051	.056	.041	-.030	-.048	.028	.024	.010	-.055
.900	.089	.078	.079	.061	-.049	-.018	.040	.031	.007	-.084

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(c) $\alpha = -9.7°$; M = 0.85

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.872	.893	.747	.466	.047	.690	.873	.765	.490	.033
.025	.237	-.115	-.549	-.860	-1.405	.311	-.095	-1.014	-1.426	-1.021
.075	.101	-.142	-.435	-.801	-1.075	.119	-.176	-.409	-1.291	-1.014
.150	-.019	-.207	-.395	-.676	-.906	.015	-.195	-.392	-1.158	-.898
.250	-.049	-.168	-.255	-.346	-.635	-.066	-.213	-.326	-.305	-.824
.350	-.084	-.162	-.223	-.239	-.455	-.136	-.239	-.306	-.259	-.703
.450	-.136	-.182	-.212	-.192	-.352	-.195	-.261	-.275	-.223	-.567
.550	-.126	-.144	-.152	-.126	-.260	-.165	-.199	-.195	-.149	-.456
.650	-.089	-.090	-.075	-.061	-.200	-.120	-.128	-.121	-.085	-.377
.750	-.044	-.036	-.021	-.002	-.124	-.054	-.047	-.035	-.012	-.290
.850	.025	.040	.049	.059	-.040	.034	.035	.050	.065	-.198
.900	.059	.075	.086	.090	.000	.076	.081	.092	.102	-.138
	Right side									
.025	-.420	-.017	.283	.534	.764	-.953	-.077	.330	.565	.750
.075	-.450	-.151	.079	.274	.481	-.542	-.166	.105	.303	.494
.150	-.396	-.186	-.018	.199	.308	-.384	-.185	.003	.160	.322
.250	-.268	-.152	-.042	.066	.190	-.308	-.203	-.074	.048	.183
.350	-.234	-.173	-.102	-.017	.084	-.279	-.226	-.139	-.044	.069
.450	-.211	-.192	-.154	-.091	-.017	-.246	-.227	-.182	-.116	-.031
.550	-.140	-.149	-.138	-.097	-.053	-.185	-.190	-.172	-.129	-.081
.650	-.069	-.088	-.092	-.071	-.058	-.100	-.114	-.128	-.091	-.081
.750	-.010	-.036	-.045	-.037	-.050	-.013	-.027	-.029	-.032	-.058
.850	.069	.051	.038	.039	.003	.074	.058	.055	.050	-.016
.900	.095	.078	.066	.060	.014	.101	.078	.063	.062	-.031
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.671	.879	.758	.473	.092	.727	.859	.695	.416	.102
.025	.338	-.124	-1.134	-1.495	-.855	.199	-.259	-1.082	-1.463	-.590
.075	.136	-.189	-.855	-1.343	-.868	-.002	-.243	-.870	-1.398	-.595
.150	.004	-.225	-.442	-1.228	-.729	-.086	-.205	-.266	-.897	-.611
.250	-.090	-.242	-.323	-.757	-.686	-.039	-.131	-.238	-.459	-.597
.350	-.154	-.249	-.301	-.185	-.652					
.450	-.194	-.249	-.278	-.164	-.604					
.550	-.178	-.212	-.216	-.155	-.534					
.650	-.131	-.156	-.136	-.102	-.453	-.113	-.129	-.189	-.385	-.418
.750	-.052	-.044	-.051	-.038	-.369	-.059	-.065	-.144	-.407	-.381
.850	.021	.017	.020	.022	-.291	-.016	-.014	-.122	-.472	-.341
.900	.044	.048	.055	.058	-.261	.000	.013	-.087	-.453	-.325
	Right side									
.025	-1.106	-.088	.360	.587	.754	-1.099	-.183	.234	.437	.595
.075	-.820	-.185	.112	.314	.496	-.730	-.250	-.045	.105	.275
.150	-.377	-.199	.006	.162	.324	-.239	-.188	-.079	-.022	.119
.250	-.320	-.236	-.102	.029	.168					
.350	-.299	-.247	-.168	-.074	.037					
.450	-.262	-.253	-.205	-.134	-.053					
.550	-.200	-.212	-.186	-.138	-.095					
.650	-.116	-.125	-.136	-.094	-.091	-.160	-.107	-.095	-.077	-.090
.750	-.035	-.040	-.041	-.039	-.082	-.164	-.057	-.062	-.055	-.102
.850	.058	.051	.042	.043	-.050	-.091	.021	.009	.009	-.071
.900	.089	.082	.070	.065	-.075	-.069	.033	.020	.006	-.101

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continued

(d) $\alpha = -9.8°$; $M = 0.90$

$\dfrac{x}{c_v}$	C_p for – , $z/b_v = 0.11$					C_p for – , $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.885	.917	.801	.564	.167	.725	.887	.816	.588	.158
.025	.203	-.105	-.479	-.811	-1.258	.274	-.089	-.898	-1.226	-.855
.075	.067	-.137	-.464	-.746	-.985	.078	-.183	-.681	-1.133	-.858
.150	-.056	-.218	-.435	-.689	-.854	-.027	-.196	-.487	-1.067	-.833
.250	-.091	-.182	-.314	-.653	-.609	-.107	-.226	-.353	-.930	-.779
.350	-.125	-.180	-.252	-.352	-.529	-.205	-.275	-.376	-.522	-.668
.450	-.190	-.216	-.242	-.249	-.458	-.284	-.321	-.286	-.197	-.569
.550	-.186	-.174	-.154	-.119	-.362	-.244	-.227	-.187	-.099	-.469
.650	-.141	-.100	-.079	-.046	-.287	-.178	-.136	-.112	-.044	-.416
.750	-.080	-.036	-.011	.013	-.218	-.086	-.047	-.028	.015	-.361
.850	-.005	.044	.061	.072	-.133	.005	.044	.057	.083	-.289
.900	.038	.080	.095	.096	-.085	.055	.088	.102	.123	-.235
Right side										
.025	-.390	-.024	.284	.529	.778	-.871	-.088	.312	.551	.752
.075	-.515	-.167	.072	.274	.496	-.714	-.192	.089	.292	.498
.150	-.476	-.214	-.026	.131	.316	-.576	-.210	-.012	.149	.325
.250	-.356	-.170	-.053	.055	.198	-.351	-.228	-.093	.029	.178
.350	-.289	-.198	-.122	-.032	.076	-.381	-.272	-.177	-.070	.056
.450	-.260	-.226	-.184	-.118	-.037	-.272	-.281	-.235	-.157	-.065
.550	-.169	-.174	-.176	-.137	-.088	-.204	-.228	-.222	-.182	-.124
.650	-.090	-.093	-.117	-.106	-.103	-.121	-.111	-.167	-.138	-.139
.750	-.024	-.031	-.053	-.060	-.108	-.027	-.028	-.020	-.052	-.115
.850	.052	.056	.039	.028	-.053	.056	.068	.055	.044	-.075
.900	.079	.087	.064	.056	-.049	.086	.086	.065	.060	-.097
\multicolumn{6}{c}{$z/b_v = 0.66$}					\multicolumn{5}{c}{$z/b_v = 0.93$}				
Left side										
.000	.701	.895	.811	.572	.197	.735	.869	.748	.510	.199
.025	.296	-.116	-.990	-1.291	-.920	.157	-.259	-.955	-1.280	-.592
.075	.095	-.196	-.854	-1.162	-.881	-.080	-.327	-.960	-1.237	-.589
.150	-.040	-.248	-.738	-1.130	-.583	-.192	-.251	-.735	-1.029	-.593
.250	-.145	-.286	-.453	-1.058	-.569	-.095	-.129	-.278	-.945	-.560
.350	-.237	-.303	-.286	-.823	-.583					
.450	-.293	-.309	-.231	-.367	-.581					
.550	-.255	-.252	-.216	-.151	-.559					
.650	-.189	-.125	-.134	-.040	-.517	-.153	-.136	-.185	-.351	-.425
.750	-.074	-.048	-.043	.019	-.453	-.087	-.067	-.181	-.416	-.399
.850	-.009	.028	.028	.060	-.373	-.037	-.009	-.187	-.442	-.365
.900	.023	.059	.065	.082	-.342	-.017	.020	-.172	-.406	-.358
Right side										
.025	-.977	-.099	.337	.569	.752	-.973	-.202	.206	.420	.590
.075	-.864	-.211	.093	.298	.491	-1.000	-.354	-.109	.080	.270
.150	-.724	-.230	-.016	.150	.320	-.733	-.243	-.164	-.060	.103
.250	-.477	-.294	-.136	.001	.158					
.350	-.296	-.306	-.226	-.115	.011					
.450	-.253	-.302	-.278	-.204	-.099					
.550	-.225	-.250	-.238	-.214	-.156					
.650	-.139	-.104	-.168	-.142	-.156	-.216	-.119	-.112	-.098	-.141
.750	-.053	-.041	-.049	-.059	-.144	-.260	-.061	-.070	-.066	-.152
.850	.042	.062	.048	.036	-.095	-.204	.026	.014	.004	-.105
.900	.078	.094	.073	.060	-.127	-.197	.042	.025	.004	-.133

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(e) $\alpha = -9.8°$; $M = 0.92$

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.919	.929	.811	.580	.101	.755	.884	.819	.601	.092
.025	.230	-.109	-.469	-.851	-1.456	.295	-.098	-.864	-1.172	-1.039
.075	.091	-.147	-.492	-.749	-1.184	.093	-.203	-.733	-1.090	-1.049
.150	-.040	-.237	-.471	-.724	-1.057	-.018	-.239	-.588	-1.049	-1.050
.250	-.080	-.214	-.509	-.734	-.801	-.101	-.253	-.553	-.976	-.972
.350	-.120	-.210	-.302	-.510	-.712	-.194	-.305	-.424	-.762	-.839
.450	-.195	-.254	-.285	-.382	-.626	-.300	-.367	-.410	-.545	-.722
.550	-.240	-.261	-.188	-.193	-.535	-.329	-.358	-.166	-.100	-.646
.650	-.172	-.138	-.079	-.063	-.470	-.216	-.143	-.081	-.029	-.630
.750	-.080	-.044	-.005	.009	-.417	-.066	-.045	-.010	.026	-.587
.850	.008	.032	.061	.066	-.341	.026	.041	.064	.086	-.516
.900	.050	.071	.098	.095	-.280	.075	.089	.108	.122	-.436
Right side										
.025	-.365	-.031	.295	.544	.743	-.822	-.105	.318	.551	.708
.075	-.522	-.186	.078	.283	.428	-.731	-.220	.093	.290	.422
.150	-.494	-.241	-.031	.135	.230	-.616	-.245	-.015	.148	.233
.250	-.507	-.197	-.058	.055	.095	-.563	-.241	-.087	.032	.074
.350	-.332	-.223	-.128	-.042	-.043	-.417	-.308	-.180	-.079	-.066
.450	-.316	-.274	-.201	-.141	-.178	-.423	-.370	-.279	-.199	-.211
.550	-.185	-.252	-.223	-.180	-.246	-.168	-.338	-.307	-.250	-.299
.650	-.070	-.122	-.155	-.162	-.280	-.074	-.120	-.201	-.210	-.336
.750	-.003	-.044	-.068	-.096	-.302	.004	-.028	-.035	-.077	-.305
.850	.073	.046	.030	.009	-.230	.083	.062	.053	.026	-.246
.900	.103	.080	.062	.040	-.224	.110	.085	.072	.051	-.270
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.732	.892	.819	.591	.143	.764	.864	.757	.528	.133
.025	.309	-.124	-.938	-1.229	-1.066	.167	-.266	-.896	-1.187	-.807
.075	.107	-.215	-.827	-1.115	-1.004	-.076	-.362	-.920	-1.178	-.755
.150	-.034	-.268	-.789	-1.087	-.721	-.216	-.328	-.760	-.993	-.757
.250	-.145	-.320	-.709	-1.069	-.712	-.114	-.179	-.648	-.888	-.716
.350	-.257	-.394	-.623	-1.008	-.722					
.450	-.348	-.397	-.273	-.637	-.723					
.550	-.363	-.345	-.104	-.389	-.715					
.650	-.295	-.132	-.085	-.195	-.693	-.119	-.141	-.151	-.605	-.594
.750	-.031	-.032	-.022	-.041	-.640	-.057	-.069	-.141	-.503	-.577
.850	.015	.020	.040	.051	-.575	-.013	-.013	-.176	-.422	-.542
.900	.044	.051	.071	.081	-.532	.004	.015	-.189	-.376	-.525
Right side										
.025	-.906	-.116	.334	.557	.700	-.897	-.207	.206	.412	.524
.075	-.823	-.232	.092	.291	.416	-.944	-.419	-.124	.061	.163
.150	-.763	-.253	-.018	.143	.228	-.756	-.316	-.201	-.083	-.024
.250	-.730	-.328	-.145	-.012	.042					
.350	-.662	-.413	-.253	-.144	-.121					
.450	-.296	-.394	-.337	-.266	-.256					
.550	-.097	-.347	-.349	-.313	-.342					
.650	-.068	-.075	-.192	-.235	-.352	-.126	-.113	-.090	-.146	-.313
.750	-.014	-.036	-.032	-.077	-.332	-.149	-.068	-.053	-.090	-.331
.850	.070	.058	.054	.021	-.253	-.153	.015	.023	-.016	-.264
.900	.100	.091	.081	.048	-.286	-.174	.033	.032	-.023	-.290

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(f) $\alpha = 0°$; M = 0.60

$\frac{x}{c_v}$	C_p for - $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for - $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.615	.643	.345	-.487	-1.393	.311	.736	.324	-.450	-1.181
.025	.269	-.092	-.520	-1.130	-2.260	.329	-.090	-.625	-1.156	-1.137
.075	.133	-.111	-.369	-.683	-1.607	.149	-.129	-.430	-1.042	-1.107
.150	.036	-.136	-.304	-.478	-.683	.052	-.141	-.316	-.776	-1.051
.250	.024	-.097	-.200	-.293	-.416	-.013	-.150	-.262	-.422	-1.007
.350	-.015	-.104	-.169	-.231	-.339	-.068	-.162	-.246	-.307	-.865
.450	-.054	-.113	-.155	-.193	-.286	-.094	-.164	-.223	-.247	-.655
.550	-.052	-.101	-.113	-.137	-.290	-.084	-.139	-.160	-.184	-.490
.650	-.034	-.064	-.067	-.091	-.200	-.066	-.104	-.107	-.121	-.362
.750	-.001	-.016	-.009	-.042	-.116	-.020	-.029	-.039	-.051	-.239
.850	.040	.033	.035	.014	-.023	.040	.029	.024	.026	-.125
.900	.070	.068	.061	.042	.014	.073	.063	.061	.061	-.060
Right side										
.025	-.442	-.029	.319	.609	.847	-.613	-.062	.377	.606	.745
.075	-.364	-.108	.131	.348	.563	-.408	-.120	.152	.362	.528
.150	-.290	-.127	.038	.208	.377	-.313	-.134	.054	.217	.373
.250	-.197	-.101	.010	.133	.268	-.244	-.143	-.009	.121	.242
.350	-.167	-.108	-.030	.066	.173	-.225	-.155	-.060	.042	.154
.450	-.151	-.127	-.074	.005	.093	-.121	-.101	-.074	-.051	-.025
.550	-.103	-.099	-.055	-.002	.059	-.158	-.134	-.086	-.023	.033
.650	-.052	-.060	-.039	.007	.054	-.096	-.090	-.060	-.011	.026
.750	-.006	-.023	-.023	.010	.042	-.027	-.029	-.013	.017	.028
.850	.052	.042	.035	.059	.070	.047	.042	.040	.061	.045
.900	1.099	1.102	1.104	1.102	1.109	.073	.061	.059	.066	.038
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.225	.713	.261	-.410	-.672	.415	.681	.228	-.214	-.472
.025	.343	-.122	-.830	-1.142	-.732	.211	-.215	-.739	-.748	-.509
.075	.158	-.145	-.483	-1.098	-.711	.036	-.159	-.379	-.793	-.523
.150	.043	-.159	-.367	-.927	-.697	-.050	-.150	-.279	-.685	-.579
.250	-.024	-.162	-.283	-.599	-.669	-.096	-.155	-.234	-.515	-.562
.350	-.077	-.169	-.244	-.382	-.641					
.450	-.105	-.173	-.223	-.270	-.604					
.550	-.105	-.155	-.183	-.205	-.562					
.650	-.068	-.106	-.118	-.137	-.513	-.087	-.085	-.116	-.317	-.386
.750	-.034	-.041	-.051	-.072	-.441	-.038	-.032	-.074	-.289	-.362
.850	.020	.015	.012	-.016	-.372	-.013	.010	-.046	-.254	-.337
.900	.043	.033	.031	.007	-.341	.003	.022	-.037	-.224	-.323
Right side										
.025	-.682	-.074	.393	.595	.700	-.611	-.132	.261	.408	.498
.075	-.465	-.136	.152	.348	.491	-.355	-.162	.024	.135	.247
.150	-.318	-.141	.056	.210	.342	-.262	-.150	-.048	.014	.098
.250	-.271	-.157	-.027	.100	.212					
.350	-.244	-.169	-.081	.019	.117					
.450	-.207	-.166	-.104	-.028	.042					
.550	-.161	-.143	-.097	-.037	.010					
.650	-.105	-.097	-.069	-.025	-.011	-.098	-.078	-.081	-.079	-.107
.750	-.043	-.048	-.030	-.011	-.032	-.068	-.034	-.060	-.067	-.113
.850	.040	.031	.040	.042	-.023	-.015	.019	-.011	-.028	-.109
.900	.066	.059	.061	.054	-.051	-.006	.031	.000	-.032	-.134

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(g) $\alpha = 0°$; M = 0.80

$\frac{x}{c_v}$	C_p for —					C_p for —				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.713	.667	.534	.177	-.326	.484	.773	.533	.044	-.579
.025	.284	-.068	-.527	-1.160	-1.538	.330	-.081	-.931	-1.345	-.961
.075	.144	-.094	-.361	-.890	-1.260	.154	-.129	-.440	-1.131	-.969
.150	.043	-.140	-.315	-.505	-.978	.056	-.144	-.330	-.871	-.925
.250	.023	-.100	-.201	-.302	-.639	-.018	-.155	-.269	-.486	-.848
.350	-.024	-.115	-.164	-.236	-.408	-.076	-.177	-.247	-.311	-.749
.450	-.075	-.134	-.152	-.197	-.325	-.116	-.187	-.219	-.253	-.631
.550	-.070	-.106	-.102	-.136	-.283	-.105	-.152	-.152	-.180	-.515
.650	-.050	-.065	-.053	-.082	-.187	-.085	-.108	-.090	-.113	-.422
.750	-.019	-.017	.013	-.026	-.128	-.027	-.022	-.014	-.031	-.326
.850	.036	.048	.070	.034	-.051	.043	.046	.063	.050	-.223
.900	.072	.083	.095	.070	-.006	.085	.089	.097	.095	-.158
Right side										
.025	-.401	.015	.345	.606	.828	-.931	-.037	.379	.603	.749
.075	-.379	-.089	.158	.358	.562	-.428	-.118	.172	.367	.537
.150	-.310	-.118	.066	.224	.390	-.327	-.128	.072	.230	.383
.250	-.209	-.098	.029	.147	.280	-.263	-.144	.004	.124	.251
.350	-.182	-.118	-.022	.072	.170	-.235	-.161	-.053	.046	.149
.450	-.163	-.134	-.068	.004	.084	-.128	-.100	-.067	-.046	-.028
.550	-.105	-.104	-.058	-.005	.054	-.160	-.143	-.090	-.037	.022
.650	-.050	-.055	-.034	-.002	.034	-.090	-.094	-.056	-.023	.005
.750	.005	-.015	-.008	.010	.016	-.007	-.009	.001	.017	.011
.850	.069	.058	.053	.069	.048	.074	.066	.066	.075	.023
.900	1.164	1.166	1.167	1.166	1.160	.095	.086	.087	.087	.006
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.422	.748	.473	.060	-.404	.501	.704	.395	.075	-.286
.025	.337	-.118	-1.061	-.919	-.695	.209	-.241	-1.001	-.620	-.453
.075	.155	-.152	-.555	-.882	-.678	.008	-.184	-.555	-.611	-.455
.150	.039	-.171	-.383	-.825	-.664	-.073	-.163	-.276	-.606	-.514
.250	-.038	-.175	-.290	-.746	-.633	-.119	-.171	-.232	-.533	-.520
.350	-.097	-.186	-.246	-.613	-.602					
.450	-.136	-.189	-.218	-.448	-.565					
.550	-.127	-.166	-.170	-.308	-.518					
.650	-.091	-.104	-.093	-.188	-.481	-.084	-.066	-.102	-.308	-.364
.750	-.039	-.023	-.030	-.088	-.429	-.032	-.011	-.051	-.276	-.342
.850	.043	.034	.040	-.014	-.371	.002	.037	-.024	-.237	-.314
.900	.050	.061	.067	.020	-.345	.023	.055	-.013	-.211	-.299
Right side										
.025	-1.041	-.046	.395	.595	.715	-.958	-.134	.264	.415	.528
.075	-.497	-.138	.167	.352	.513	-.439	-.184	.007	.135	.268
.150	-.343	-.138	.073	.221	.359	-.264	-.163	-.064	-.002	.102
.250	-.298	-.171	-.021	.103	.228					
.350	-.257	-.181	-.081	.014	.109					
.450	-.214	-.180	-.113	-.045	.028					
.550	-.160	-.143	-.105	-.056	-.017					
.650	-.093	-.092	-.068	-.036	-.031	-.090	-.066	-.061	-.074	-.119
.750	-.021	-.023	-.014	-.011	-.049	-.048	-.012	-.039	-.054	-.121
.850	.063	.066	.067	.054	-.037	.013	.055	.032	-.014	-.102
.900	.099	.095	.087	.070	-.068	.023	.069	.038	-.016	-.128

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(h) α = 0°; M = 0.85

$\frac{x}{c_v}$	C_p for -					C_p for -				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.743	.682	.589	.076	-.137	.538	.787	.593	-.077	-.387
.025	.279	-.057	-.529	-1.682	-1.311	.328	-.076	-1.084	-2.209	-.876
.075	.143	-.088	-.385	-1.402	-1.104	.148	-.124	-.434	-2.014	-.892
.150	.042	-.136	-.340	-1.120	-.920	.049	-.147	-.356	-1.734	-.882
.250	.020	-.096	-.220	-.703	-.711	-.024	-.160	-.298	-.694	-.808
.350	-.037	-.121	-.194	-.601	-.517	-.093	-.181	-.279	-.660	-.721
.450	-.096	-.144	-.181	-.545	-.408	-.142	-.196	-.252	-.630	-.616
.550	-.089	-.108	-.119	-.473	-.321	-.131	-.161	-.174	-.538	-.516
.650	-.069	-.073	-.066	-.411	-.209	-.105	-.113	-.102	-.437	-.434
.750	-.030	-.013	.002	-.346	-.140	-.043	-.017	-.029	-.333	-.341
.850	.028	.056	.062	-.254	-.057	.043	.060	.056	-.231	-.240
.900	.062	.092	.093	-.211	-.011	.078	.097	.096	-.183	-.177
	Right side									
.025	-.384	.019	.339	.455	.833	-.971	-.037	.366	.435	.750
.075	-.398	-.086	.142	.141	.560	-.431	-.114	.155	.139	.537
.150	-.326	-.111	.060	-.019	.395	-.344	-.131	.054	-.028	.381
.250	-.223	-.098	.020	-.120	.281	-.279	-.147	-.018	-.161	.260
.350	-.201	-.126	-.044	-.224	.174	-.257	-.174	-.080	-.260	.151
.450	-.174	-.143	-.093	-.315	.080	-.135	-.100	-.080	-.373	-.036
.550	-.115	-.106	-.083	-.330	.047	-.165	-.147	-.119	-.378	.004
.650	-.050	-.061	-.058	-.319	.022	-.088	-.088	-.086	-.355	-.016
.750	.003	-.011	-.029	-.308	.009	.000	-.007	-.019	-.296	-.009
.850	.071	.066	.044	-.227	.042	.081	.074	.063	-.220	.014
.900	1.187	1.192	1.192	1.182	1.184	.110	.103	.079	-.202	-.002
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.477	.763	.533	-.150	-.306	.531	.720	.442	-.245	-.243
.025	.340	-.110	-1.180	-2.279	-.689	.200	-.256	-1.087	-1.328	-.477
.075	.156	-.153	-.560	-2.093	-.698	-.014	-.204	-.686	-1.283	-.490
.150	.035	-.178	-.411	-1.493	-.702	-.098	-.177	-.297	-1.265	-.557
.250	-.050	-.190	-.316	-1.209	-.658	-.139	-.191	-.253	-1.084	-.543
.350	-.113	-.204	-.268	-.928	-.629					
.450	-.159	-.207	-.240	-.674	-.577					
.550	-.151	-.170	-.187	-.534	-.517					
.650	-.103	-.110	-.108	-.425	-.465	-.096	-.066	-.121	-.728	-.385
.750	-.054	-.011	-.029	-.335	-.408	-.036	-.003	-.069	-.671	-.359
.850	.041	.049	.041	-.258	-.349	.010	.042	-.037	-.610	-.328
.900	.055	.069	.070	-.227	-.319	.026	.069	-.021	-.570	-.315
	Right side									
.025	-1.149	-.050	.382	.428	.712	-1.108	-.150	.242	.223	.530
.075	-.436	-.137	.153	.133	.512	-.470	-.204	-.032	-.154	.243
.150	-.362	-.143	.051	-.037	.359	-.264	-.178	-.102	-.349	.088
.250	-.303	-.180	-.044	-.190	.220					
.350	-.266	-.196	-.113	-.314	.103					
.450	-.227	-.191	-.148	-.391	.018					
.550	-.168	-.158	-.142	-.411	-.036					
.650	-.093	-.096	-.090	-.382	-.051	-.089	-.070	-.077	-.427	-.141
.750	-.016	-.013	-.031	-.330	-.062	-.044	-.007	-.050	-.389	-.135
.850	.078	.079	.062	-.229	-.042	.019	.072	.021	-.321	-.111
.900	.111	.107	.083	-.208	-.069	.030	.083	.036	-.321	-.134

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(i) $\alpha = 0°$; M = 0.90

$\dfrac{x}{C_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.775	.693	.645	.419	.025	.600	.792	.664	.335	-.189
.025	.284	-.050	-.462	-.939	-1.114	.318	-.084	-.944	-1.301	-.743
.075	.147	-.089	-.383	-.769	-.971	.139	-.140	-.610	-1.169	-.755
.150	.030	-.156	-.378	-.645	-.828	.039	-.162	-.395	-1.053	-.769
.250	.011	-.118	-.241	-.494	-.659	-.041	-.185	-.324	-.610	-.749
.350	-.052	-.146	-.218	-.331	-.564	-.117	-.226	-.331	-.453	-.689
.450	-.114	-.180	-.206	-.260	-.508	-.194	-.256	-.269	-.296	-.614
.550	-.121	-.149	-.135	-.143	-.425	-.177	-.202	-.174	-.106	-.534
.650	-.101	-.097	-.068	-.067	-.330	-.126	-.134	-.101	-.043	-.468
.750	-.048	-.023	.010	-.002	-.229	-.053	-.029	-.019	.018	-.394
.850	.021	.050	.069	.057	-.133	.031	.055	.065	.087	-.314
.900	.057	.087	.101	.088	-.070	.074	.099	.109	.124	-.254
Right side										
.025	-.337	.025	.336	.600	.833	-.890	-.044	.352	.581	.745
.075	-.411	-.093	.148	.361	.567	-.616	-.129	.150	.353	.532
.150	-.379	-.140	.044	.218	.396	-.403	-.154	.048	.217	.376
.250	-.260	-.122	.009	.141	.279	-.323	-.174	-.030	.108	.253
.350	-.237	-.152	-.056	.055	.166	-.324	-.213	-.100	.018	.138
.450	-.216	-.181	-.116	-.027	.063	-.200	-.177	-.131	-.065	.009
.550	-.132	-.145	-.112	-.049	.019	-.178	-.193	-.164	-.097	-.031
.650	-.061	-.085	-.080	-.040	-.004	-.094	-.117	-.109	-.073	-.056
.750	.006	-.018	-.029	-.008	-.020	-.005	-.010	-.023	-.008	-.045
.850	.078	.065	.048	.061	.008	.085	.078	.061	.067	-.025
.900	.101	.098	.069	.076	.009	.115	.106	.085	.086	-.046
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.532	.771	.602	.271	-.158	.564	.717	.493	.141	-.201
.025	.317	-.125	-1.058	-1.377	-.719	.171	-.299	-1.021	-1.325	-.487
.075	.137	-.174	-.874	-1.243	-.731	-.064	-.284	-1.013	-1.329	-.497
.150	.014	-.210	-.406	-1.150	-.734	-.197	-.226	-.437	-1.079	-.557
.250	-.081	-.232	-.407	-1.090	-.636	-.188	-.232	-.178	-.741	-.557
.350	-.166	-.260	-.264	-.682	-.606					
.450	-.217	-.258	-.226	-.124	-.586					
.550	-.198	-.214	-.191	-.064	-.556					
.650	-.129	-.121	-.105	-.024	-.515	-.109	-.076	-.112	-.261	-.411
.750	-.050	-.025	-.021	.018	-.463	-.038	-.009	-.050	-.255	-.390
.850	.026	.043	.050	.063	-.399	-.001	.041	-.027	-.233	-.362
.900	.046	.073	.084	.084	-.372	.018	.066	-.015	-.203	-.346
Right side										
.025	-1.017	-.058	.360	.572	.709	-.997	-.174	.219	.410	.528
.075	-.810	-.156	.140	.337	.505	-.955	-.293	-.077	.098	.261
.150	-.387	-.160	.040	.205	.355	-.377	-.230	-.190	-.088	.069
.250	-.404	-.221	-.068	.074	.213					
.350	-.313	-.152	-.152	-.040	.080					
.450	-.225	-.242	-.197	-.116	-.016					
.550	-.185	-.194	-.172	-.131	-.076					
.650	-.100	-.109	-.108	-.090	-.094	.055	.001	-.022	-.049	-.097
.750	-.019	-.025	-.040	-.028	-.102	-.050	-.010	-.049	-.051	-.166
.850	.079	.077	.056	.062	-.070	.018	.062	.029	.008	-.134
.900	.110	.113	.089	.090	-.097	.031	.083	.042	.018	-.152

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(j) $\alpha = 0°$; M = 0.92

$\frac{x}{c_v}$	C_p for $z/b_v = 0.11$					C_p for $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.781	.698	.658	.446	.072	.613	.802	.681	.366	-.142
.025	.286	-.036	-.440		-1.092	.317	-.067	-.912	-1.253	-.726
.075	.152	-.079	-.375	-.754	-.950	.140	-.136	-.687	-1.136	-.731
.150	.034	-.149	-.367	-.645	-.815	.040	-.159	-.414	-1.036	-.742
.250	.009	-.114	-.257	-.549	-.648	-.039	-.181	-.324	-.674	-.730
.350	-.053	-.146	-.224	-.373	-.558	-.119	-.229	-.345	-.520	-.684
.450	-.129	-.185	-.223	-.311	-.514	-.201	-.268	-.332	-.420	-.619
.550	-.134	-.156	-.141	-.175	-.441	-.197	-.213	-.159	-.120	-.545
.650	-.106	-.095	-.064	-.077	-.354	-.142	-.140	-.082	-.032	-.477
.750	-.053	-.020	.012	-.005	-.249	-.051	-.020	-.006	.030	-.405
.850	.023	.061	.080	.059	-.151	.035	.065	.075	.097	-.325
.900	.060	.097	.109	.093	-.095	.082	.110	.117	.125	-.273
Right side										
.025	-.324	.036	.343	.612	.835	-.876	-.035	.351	.582	.747
.075	-.410	-.084	.149	.364	.566	-.645	-.123	.149	.353	.532
.150	-.379	-.132	.054	.224	.396	-.427	-.149	.054	.214	.375
.250	-.271	-.119	.011	.147	.282	-.321	-.172	-.026	.104	.249
.350	-.241	-.149	-.054	.054	.168	-.338	-.211	-.098	.014	.139
.450	-.226	-.183	-.122	-.037	.059	-.250	-.215	-.153	-.077	.017
.550	-.136	-.149	-.124	-.058	.017	-.163	-.200	-.183	-.116	-.036
.650	-.055	-.078	-.087	-.051	-.019	-.080	-.109	-.119	-.096	-.068
.750	.014	-.012	-.034	-.026	-.037	.008	-.007	-.024	-.019	-.058
.850	.078	.074	.049	.053	-.004	.092	.087	.068	.064	-.037
.900	.109	.102	.074	.074	-.015	.120	.114	.095	.083	-.062

$\frac{x}{c_v}$	C_p for $z/b_v = 0.66$					C_p for $z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.542	.772	.620	.302	-.117	.571	.724	.518	.185	-.176
.025	.321	-.110	-1.014	-1.325	-.737	.170	-.275	-.967	-1.282	-.486
.075	.140	-.165	-.864	-1.200	-.751	-.063	-.298	-.984	-1.282	-.509
.150	.018	-.205	-.602	-1.112	-.728	-.218	-.247	-.738	-1.064	-.562
.250	-.081	-.233	-.390	-1.071	-.699	-.244	-.242	-.185	-.917	-.566
.350	-.177	-.295	-.413	-.962	-.611					
.450	-.247	-.273	-.172	-.358	-.583					
.550	-.209	-.219	-.167	-.043	-.558					
.650	-.131	-.106	-.094	.006	-.526	-.092	-.065	-.111	-.311	-.427
.750	-.042	-.016	-.020	.044	-.473	-.034	-.001	-.051	-.274	-.400
.850	.033	.058	.061	.078	-.413	.006	.053	-.023	-.244	-.370
.900	.055	.087	.091	.098	-.391	.026	.078	-.011	-.219	-.361
Right side										
.025	-.995	-.053	.361	.568	.710	-.983	-.167	.220	.411	.536
.075	-.826	-.146	.139	.339	.503	-.960	-.316	-.079	.101	.261
.150	-.510	-.158	.041	.202	.352	-.675	-.251	-.215	-.095	.067
.250	-.397	-.222	-.071	.067	.208					
.350	-.421	-.293	-.170	-.048	.080					
.450	-.181	-.251	-.231	-.138	-.025					
.550	-.164	-.189	-.185	-.170	-.092					
.650	-.087	-.096	-.111	-.113	-.112	-.083	-.058	-.069	-.086	-.150
.750	-.011	-.016	-.032	-.037	-.121	-.055	-.004	-.035	-.052	-.181
.850	.086	.089	.070	.063	-.075	.018	.076	.038	.014	-.144
.900	.120	.123	.104	.091	-.106	.030	.093	.057	.017	-.164

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(k) $\alpha = 9.4°$; $M = 0.60$

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.404	.804	.439	-.646	-1.194	.162	.693	.174	-.716	-1.505
.025	-.031	-.132	-.231	-.734	-2.578	.333	-.109	-.701	-1.213	-1.315
.075	.015	-.136	-.316	-.535	-.893	.149	-.141	-.463	-1.045	-1.265
.150	-.019	-.164	-.314	-.444	-.664	.054	-.143	-.341	-.793	-1.333
.250	-.019	-.109	-.225	-.319	-.427	-.012	-.155	-.282	-.439	-1.176
.350	-.044	-.116	-.206	-.265	-.359	-.065	-.166	-.259	-.317	-.806
.450	-.065	-.132	-.195	-.231	-.345	-.095	-.180	-.234	-.274	-.457
.550	-.067	-.116	-.158	-.190	-.258	-.093	-.141	-.179	-.213	-.308
.650	-.044	-.079	-.117	-.149	-.180	-.070	-.104	-.126	-.154	-.240
.750	-.015	-.038	-.062	-.074	-.114	-.035	-.063	-.062	-.086	-.164
.850	.015	.011	.002	-.013	-.037	.020	.008	.013	-.009	-.078
.900	.041	.045	.039	.016	.007	.045	.045	.045	.025	-.034
Right side										
.025	-.178	-.044	.000	.504	.806	-.684	-.063	.373	.583	.715
.075	-.327	-.136	.013	.320	.543	-.451	-.116	.162	.363	.536
.150	-.309	-.146	-.016	.214	.383	-.343	-.129	.064	.227	.383
.250	-.238	-.102	-.028	.150	.281	-.281	-.141	.004	.134	.267
.350	-.221	-.118	-.048	.073	.180	-.256	-.157	-.046	.062	.178
.450	-.203	-.132	-.074	.019	.107	-.231	-.157	-.078	.003	.100
.550	-.159	-.107	-.060	.005	.073	-.194	-.136	-.080	-.020	.059
.650	-.109	-.072	-.037	.010	.062	-.125	-.093	-.051	-.013	.039
.750	-.056	-.024	-.005	.019	.057	-.054	-.031	-.014	.016	.048
.850	.011	.041	.039	.062	.087	.025	.038	.050	.057	.078
.900	.045	.061	.064	.071	.082	.054	.050	.055	.053	.055
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.041	.643	.034	-.841	-1.110	.188	.585	-.039	-.485	-.811
.025	.337	-.139	-.977	-1.369	-.897	.199	-.226	-.872	-.730	-.594
.075	.156	-.159	-.517	-1.340	-.884	.006	-.187	-.453	-.711	-.587
.150	.043	-.164	-.387	-1.176	-.849	-.088	-.173	-.286	-.673	-.594
.250	-.026	-.164	-.300	-.639	-.813	-.079	-.104	-.197	-.573	-.578
.350	-.074	-.173	-.261	-.315	-.781					
.450	-.113	-.178	-.227	-.242	-.738					
.550	-.125	-.164	-.190	-.210	-.664					
.650	-.090	-.109	-.131	-.151	-.575	-.111	-.072	-.122	-.340	-.461
.750	-.044	-.051	-.064	-.086	-.480	-.072	-.024	-.087	-.297	-.434
.850	-.003	-.003	-.007	-.036	-.379	-.035	.013	-.058	-.253	-.416
.900	.015	.025	.018	-.011	-.336	-.019	.027	-.037	-.231	-.395
Right side										
.025	-.881	-.065	.380	.570	.644	-.718	-.127	.249	.375	.450
.075	-.502	-.129	.162	.361	.495	-.417	-.178	.007	.112	.237
.150	-.350	-.132	.064	.220	.356	-.270	-.152	-.074	-.015	.087
.250	-.300	-.157	-.016	.103	.228					
.350	-.272	-.173	-.071	.023	.135					
.450	-.231	-.166	-.099	-.020	.062					
.550	-.189	-.141	-.096	-.045	.020					
.650	-.125	-.107	-.071	-.036	.000	-.097	-.063	-.085	-.115	-.130
.750	-.061	-.031	-.030	-.024	-.012	-.074	-.024	-.071	-.106	-.151
.850	.015	.031	.032	.030	.000	-.026	.025	-.009	-.070	-.149
.900	.045	.054	.055	.044	-.021	-.003	.048	.004	-.056	-.171

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(*l*) $\alpha = 9.6°$; $M = 0.80$

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.356	.848	.422	-.349	-.188	.404	.723	.434	-.137	-.801
.025	-.043	-.119	-.153	-.597	-1.595	.328	-.106	-1.046	-1.517	-1.046
.075	.020	-.134	-.325	-.569	-1.058	.151	-.149	-.499	-1.094	-1.028
.150	-.016	-.176	-.363	-.537	-.705	.057	-.148	-.381	-.770	-.985
.250	-.022	-.121	-.256	-.374	-.485	-.015	-.163	-.317	-.490	-.930
.350	-.056	-.130	-.232	-.307	-.404	-.081	-.198	-.299	-.393	-.797
.450	-.081	-.154	-.224	-.272	-.360	-.127	-.213	-.270	-.327	-.631
.550	-.081	-.134	-.182	-.219	-.307	-.115	-.167	-.201	-.244	-.497
.650	-.059	-.095	-.130	-.169	-.233	-.091	-.121	-.140	-.169	-.397
.750	-.024	-.045	-.063	-.090	-.163	-.043	-.056	-.063	-.082	-.297
.850	.019	.020	.006	-.009	-.084	.025	.024	.023	-.001	-.183
.900	.051	.057	.045	.027	-.038	.060	.066	.064	.043	-.119
Right side										
.025	-.095	-.018	-.025	.390	.768	-1.077	-.048	.375	.589	.715
.075	-.333	-.131	.024	.312	.521	-.468	-.116	.169	.370	.532
.150	-.352	-.149	.003	.218	.375	-.367	-.136	.064	.237	.393
.250	-.262	-.110	-.020	.154	.274	-.300	-.148	.003	.137	.271
.350	-.241	-.130	-.052	.079	.172	-.276	-.169	-.055	.059	.180
.450	-.229	-.151	-.083	.005	.091	-.250	-.184	-.101	-.009	.088
.550	-.175	-.125	-.072	-.015	.056	-.198	-.161	-.112	-.044	.038
.650	-.116	-.080	-.044	-.008	.044	-.125	-.109	-.073	-.029	.021
.750	-.059	-.035	-.012	.002	.036	-.040	-.035	-.022	.002	.026
.850	.023	.041	.045	.050	.064	.045	.047	.045	.053	.045
.900	.058	.068	.067	.064	.058	.673	.066	.061	.059	.021
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.284	.668	.295	-.272	-.711	.321	.600	.178	-.256	-.622
.025	.336	-.140	-1.197	-1.250	-.711	.204	-.259	-.895	-.685	-.541
.075	.155	-.164	-.650	-1.173	-.714	-.015	-.232	-.680	-.664	-.523
.150	.043	-.182	-.439	-1.043	-.700	-.122	-.191	-.354	-.671	-.523
.250	-.036	-.188	-.325	-.871	-.671	-.086	-.115	-.237	-.600	-.535
.350	-.095	-.204	-.285	-.619	-.650					
.450	-.136	-.207	-.250	-.387	-.627					
.550	-.144	-.179	-.203	-.248	-.590					
.650	-.103	-.118	-.131	-.149	-.552	-.106	-.069	-.115	-.358	-.445
.750	-.043	-.044	-.054	-.062	-.500	-.053	-.012	-.072	-.310	-.422
.850	.010	.018	.012	.000	-.445	-.010	.033	-.037	-.265	-.392
.900	.037	.047	.042	.029	-.412	.004	.053	-.017	-.233	-.381
Right side										
.025	-1.103	-.059	.379	.574	.666	-.943	-.134	.256	.406	.483
.075	-.572	-.134	.166	.364	.503	-.578	-.214	-.022	.134	.263
.150	-.383	-.142	.071	.231	.366	-.277	-.167	-.096	-.015	.105
.250	-.323	-.170	-.022	.111	.238					
.350	-.283	-.195	-.086	.023	.132					
.450	-.241	-.190	-.116	-.037	.052					
.550	-.189	-.164	-.125	-.062	-.003					
.650	-.121	-.109	-.086	-.050	-.026	-.089	-.054	-.081	-.113	-.154
.750	-.045	-.039	-.031	-.023	-.043	-.059	-.015	-.052	-.096	-.169
.850	.043	.048	.045	.037	-.032	-.001	.051	.012	-.049	-.159
.900	.076	.077	.070	.055	-.064	.022	.069	.030	-.050	-.180

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(m) $\alpha = 9.7°$; M = 0.85

$\dfrac{x}{C_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.336	.866	.415	-.203	-.029	.470	.740	.511	.018	-.622
.025	-.065	-.108	-.103	-.519	-1.411	.325	-.100	-1.153	-1.579	-1.016
.075	.016	-.129	-.314	-.539	-.996	.146	-.149	-.476	-1.407	-1.032
.150	-.020	-.177	-.397	-.583	-.736	.057	-.158	-.392	-.666	-.972
.250	-.031	-.117	-.273	-.411	-.508	-.017	-.172	-.335	-.439	-.884
.350	-.066	-.134	-.247	-.318	-.433	-.088	-.209	-.317	-.382	-.767
.450	-.091	-.165	-.247	-.271	-.386	-.142	-.228	-.288	-.350	-.643
.550	-.091	-.148	-.199	-.224	-.338	-.130	-.189	-.214	-.261	-.545
.650	-.068	-.103	-.133	-.167	-.266	-.103	-.128	-.140	-.174	-.458
.750	-.037	-.050	-.070	-.093	-.200	-.054	-.059	-.062	-.085	-.365
.850	.019	.023	.005	-.004	-.117	.020	.029	.022	.009	-.245
.900	.051	.066	.046	.041	-.074	.064	.073	.071	.051	-.177
Right side										
.025	-.049	-.015	-.051	.327	.728	-1.044	-.049	.361	.586	.719
.075	-.326	-.131	.012	.294	.488	-.462	-.122	.155	.370	.532
.150	-.370	-.162	-.008	.220	.347	-.373	-.138	.062	.242	.388
.250	-.276	-.111	-.038	.165	.256	-.315	-.158	-.013	.142	.268
.350	-.255	-.136	-.068	.081	.163	-.292	-.182	-.073	.065	.169
.450	-.244	-.166	-.103	.009	.079	-.265	-.197	-.116	-.012	.076
.550	-.187	-.136	-.088	-.015	.044	-.206	-.179	-.129	-.052	.025
.650	-.120	-.088	-.059	-.009	.029	-.133	-.115	-.093	-.041	.000
.750	-.064	-.042	-.029	.002	.018	-.040	-.036	-.033	-.005	.000
.850	.024	.039	.041	.051	.041	.048	.053	.042	.055	.018
.900	.063	.070	.058	.071	.032	.081	.076	.061	.062	-.012
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.342	.680	.365	-.135	-.619	.351	.609	.207	-.240	-.613
.025	.331	-.132	-1.253	-1.365	-.702	.197	-.258	-1.163	-.764	-.545
.075	.152	-.168	-.659	-1.257	-.703	-.031	-.265	-.721	-.731	-.528
.150	.036	-.189	-.466	-1.016	-.692	-.150	-.207	-.428	-.749	-.532
.250	-.049	-.200	-.355	-.794	-.673	-.102	-.118	-.240	-.666	-.545
.350	-.113	-.220	-.296	-.613	-.650					
.450	-.154	-.223	-.264	-.412	-.631					
.550	-.170	-.194	-.213	-.264	-.598					
.650	-.119	-.121	-.136	-.146	-.564	-.110	-.073	-.125	-.347	-.474
.750	-.048	-.043	-.060	-.055	-.518	-.054	-.011	-.076	-.290	-.441
.850	.007	.022	.014	.009	-.463	-.010	.039	-.039	-.237	-.413
.900	.040	.054	.046	.045	-.439	.010	.062	-.023	-.206	-.400
Right side										
.025	-1.235	-.059	.367	.571	.671	-1.152	-.145	.242	.411	.493
.075	-.533	-.138	.155	.364	.509	-.713	-.252	-.058	.132	.268
.150	-.398	-.149	.062	.236	.372	-.289	-.177	-.130	-.032	.106
.250	-.342	-.185	-.036	.118	.240					
.350	-.302	-.214	-.102	.022	.129					
.450	-.254	-.209	-.147	-.052	.035					
.550	-.198	-.180	-.145	-.079	-.023					
.650	-.122	-.111	-.106	-.062	-.049	-.092	-.054	-.098	-.117	-.182
.750	-.042	-.037	-.049	-.028	-.069	-.059	-.012	-.066	-.098	-.195
.850	.047	.056	.039	.045	-.054	.010	.059	.011	-.045	-.182
.900	.081	.084	.066	.062	-.089	.027	.079	.025	-.043	-.205

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(n) $\alpha = 9.7°$; M = 0.90

$\frac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.300	.911	.414	-.105	.098	.531	.752	.572	.157	-.465
.025	-.092	-.081	-.019	-.446	-1.267	.324	-.088	-1.037	-1.403	-.944
.075	.017	-.109	-.248	-.485	-.927	.150	-.142	-.568	-1.266	-.955
.150	-.013	-.167	-.415	-.583	-.700	.054	-.155	-.473	-.752	-.897
.250	-.026	-.114	-.297	-.517	-.571	-.021	-.178	-.341	-.660	-.826
.350	-.066	-.141	-.265	-.421	-.488	-.092	-.224	-.361	-.504	-.755
.450	-.098	-.179	-.274	-.337	-.437	-.162	-.268	-.345	-.286	-.654
.550	-.110	-.173	-.221	-.236	-.382	-.157	-.215	-.208	-.266	-.566
.650	-.089	-.120	-.142	-.171	-.313	-.126	-.143	-.136	-.208	-.496
.750	-.049	-.056	-.068	-.109	-.238	-.065	-.060	-.054	-.105	-.416
.850	.007	.022	.009	-.022	-.149	.021	.029	.034	-.002	-.311
.900	.051	.063	.051	.027	-.096	.066	.075	.081	.046	-.235
	Right side									
.025	.038	.016	-.056	.306	.728	-.941	-.039	.377	.588	.725
.075	-.276	-.114	.033	.298	.496	-.568	-.114	.171	.368	.536
.150	-.411	-.149	.019	.229	.358	-.420	-.138	.070	.240	.394
.250	-.292	-.109	-.009	.165	.268	-.329	-.158	-.002	.133	.269
.350	-.283	-.139	-.056	.075	.167	-.341	-.203	-.069	.047	.171
.450	-.277	-.184	-.092	-.008	.076	-.305	-.230	-.133	-.040	.068
.550	-.208	-.158	-.092	-.035	.030	-.214	-.204	-.152	-.079	.006
.650	-.129	-.101	-.072	-.037	.010	-.132	-.143	-.114	-.073	-.019
.750	-.062	-.053	-.037	-.026	-.004	-.038	-.040	-.042	-.029	-.022
.850	.026	.037	.034	.035	.014	.055	.053	.042	.035	-.008
.900	.062	.066	.057	.050	.006	.085	.072	.059	.044	-.030
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.404	.684	.436	.018	-.512	.390	.616	.278	-.129	-.553
.025	.328	-.129	-1.169	-1.424	-.668	.183	-.256	-1.162	-.768	-.528
.075	.150	-.169	-.986	-1.313	-.673	-.053	-.314	-1.074	-.782	-.514
.150	.029	-.204	-.456	-1.163	-.677	-.246	-.233	-.620		-.516
.250	-.060	-.226	-.415	-.770	-.664	-.134	-.125	-.194	-.758	-.528
.350	-.138	-.265	-.393	-.612	-.644					
.450	-.200	-.263	-.229	-.509	-.620					
.550	-.197	-.218	-.206	-.397	-.589					
.650	-.134	-.128	-.130	-.243	-.559	-.118	-.076	-.122	-.399	-.475
.750	-.057	-.047	-.046	-.113	-.523	-.054	-.008	-.074	-.337	-.453
.850	.010	.023	.022	-.016	-.475	-.006	.043	-.033	-.281	-.425
.900	.041	.058	.055	.026	-.448	.014	.059	-.011	-.251	-.409
	Right side									
.025	-1.120	-.051	.370	.562	.674	-1.150	-.141	.246	.403	.502
.075	-.901	-.141	.163	.353	.509	-1.036	-.318	-.061	.119	.280
.150	-.364	-.151	.066	.226	.374	-.372	-.206	-.202	-.065	.108
.250	-.409	-.208	-.041	.099	.241					
.350	-.356	-.243	-.125	-.005	.123					
.450	-.238	-.253	-.186	-.091	.026					
.550	-.201	-.207	-.174	-.131	-.042					
.650	-.121	-.129	-.122	-.109	-.073	-.098	-.061	-.100	-.159	-.199
.750	-.044	-.041	-.053	-.060	-.087	-.061	-.016	-.064	-.125	-.213
.850	.054	.057	.041	.019	-.073	.006	.061	.015	-.068	-.193
.900	.090	.084	.070	.040	-.102	.026	.078	.030	-.064	-.208

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(o) $\alpha = 9.8°$; $M = 0.92$

$\frac{x}{c_v}$	C_p for – $z/b_v = 0.11$					C_p for – $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.313	.968	.449	-.173		.560	.749	.591	.190	
.025	-.059	-.034	.048	-.363		.329	-.038	-.978	-1.331	
.075	.044	-.063	-.164	-.408		.161	-.097	-.420	-1.189	
.150	.027	-.119	-.328	-.519		.062	-.119	-.375	-.687	
.250	.002	-.085	-.240	-.484		-.015	-.153	-.323	-.630	
.350	-.044	-.119	-.237	-.423		-.094	-.200	-.349	-.547	
.450	-.088	-.170	-.265	-.362		-.161	-.251	-.365	-.306	
.550	-.103	-.175	-.222	-.253		-.169	-.217	-.211	-.270	
.650	-.090	-.127	-.138	-.176		-.133	-.152	-.135	-.218	
.750	-.049	-.063	-.070	-.106		-.069	-.067	-.057	-.112	
.850	.010	.014	.008	-.024		.012	.023	.030	-.007	
.900	.045	.056	.052	.020		.058	.074	.076	.040	
Right side										
.025	.121	.058	.004	.233		-.855	.000	.397	.593	
.075	-.185	-.068	.071	.300		-.458	-.081	.192	.379	
.150	-.333	-.110	.063	.262		-.370	-.108	.096	.249	
.250	-.253	-.084	.033	.183		-.310	-.135	.017	.137	
.350	-.260	-.124	-.016	.083		-.340	-.188	-.057	.047	
.450	-.274	-.172	-.070	-.005		-.333	-.225	-.122	-.042	
.550	-.212	-.163	-.084	-.040		-.214	-.206	-.148	-.093	
.650	-.132	-.108	-.067	-.048		-.135	-.142	-.116	-.093	
.750	-.072	-.062	-.037	-.039		-.042	-.045	-.043	-.047	
.850	.023	.031	.030	.022		.048	.047	.039	.022	
.900	.057	.065	.056	.040		.081	.069	.056	.032	

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.414	.686	.438	.049		.413	.615	.299	-.085	
.025	.325	-.084	-1.116	-1.409		.182	-.235	-1.136	-.777	
.075	.152	-.131	-.920	-1.285		-.051	-.288	-1.063	-.772	
.150	.036	-.168	-.408	-1.132		-.245	-.222	-.506	-.789	
.250	-.057	-.204	-.416	-.764		-.144	-.115	-.177	-.755	
.350	-.143	-.258	-.425	-.620						
.450	-.208	-.269	-.223	-.517						
.550	-.204	-.217	-.197	-.406						
.650	-.139	-.133	-.131	-.258		-.123	-.078	-.125	-.402	
.750	-.066	-.049	-.051	-.124		-.056	-.007	-.072	-.338	
.850	.006	.023	.019	-.019		-.005	.041	-.030	-.279	
.900	.033	.056	.052	.024		.012	.062	-.009	-.250	
Right side										
.025	-1.052	-.030	.383	.558		-1.093	-.129	.251	.405	
.075	-.781	-.114	.181	.355		-.972	-.300	-.051	.117	
.150	-.365	-.131	.080	.227		-.349	-.189	-.189	-.063	
.250	-.399	-.191	-.029	.096						
.350	-.413	-.240	-.119	-.010						
.450	-.231	-.248	-.186	-.099						
.550	-.202	-.208	-.181	-.151		-.095	-.058	-.100	-.158	
.650	-.124	-.148	-.118	-.127		-.056	-.011	-.062	-.130	
.750	-.042	-.030	-.054	-.072		.010	.060	.016	-.069	
.850	.050	.060	.040	.012		.029	.078	.034	-.064	
.900	.083	.086	.069	.035						

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(p) $\alpha = 15.6°$; M = 0.60

$\frac{x}{C_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	-.316	.822	.197	.098	-.516	.271	.741	.372	-.419	-.959
.025	-.636	-.153	.299	.109	-.521	.212	-.108	-.367	-.717	-1.045
.075	-.184	-.155	.017	-.086	-.455	.082	-.151	-.317	-.522	-.900
.150	-.075	-.178	-.142	-.200	-.426	.041	-.153	-.275	-.401	-.718
.250	-.029	-.142	-.148	-.182	-.362	.005	-.162	-.257	-.342	-.580
.350	-.020	-.119	-.167	-.191	-.310	-.027	-.183	-.248	-.305	-.480
.450	-.020	-.130	-.181	-.191	-.272	-.057	-.189	-.229	-.269	-.394
.550	-.020	-.130	-.158	-.164	-.224	-.050	-.164	-.185	-.219	-.319
.650	-.004	-.085	-.114	-.116	-.163	-.038	-.133	-.142	-.171	-.260
.750	.000	-.056	-.077	-.073	-.106	-.018	-.078	-.084	-.114	-.183
.850	.016	-.013	-.029	-.030	-.056	.016	-.013	-.019	-.055	-.117
.900	.039	.026	-.003	.000	-.020	-.163	-.178	-.153	-.162	-.154
	Right side									
.025	.328	-.108	-.777	-.291	-.095	-.561	-.106	.128	.201	.211
.075	-.077	-.185	-.220	-.084	.019	-.402	-.160	.045	.128	.230
.150	-.202	-.167	-.072	-.002	.093	-.327	-.160	.022	.091	.230
.250	-.191	-.119	-.006	.041	.146	-.282	-.155	.017	.091	.241
.350	-.204	-.128	.008	.062	.162	-.261	-.169	.006	.080	.223
.450	-.200	-.140	.013	.064	.157	-.243	-.180	-.012	.064	.202
.550	-.168	-.121	.015	.062	.130	-.197	-.160	-.019	.055	.173
.650	-.118	-.083	.017	.057	.107	-.143	-.126	-.003	.059	.148
.750	-.077	-.049	.022	.052	.084	-.082	-.067	.015	.057	.121
.850	-.009	.010	.041	.057	.071	-.011	.005	.045	.066	.100
.900	.012	.032	.043	.057	.059	.014	.019	.036	.059	.078

$\frac{x}{C_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	-.018	.634	.077	-.633	-.936	.100	.555	-.178	-.640	-.752
.025	.378	-.128	-.976	-1.303	-.965	.209	-.216	-.959	-.802	-.612
.075	.198	-.155	-.545	-1.175	-.945	.021	-.189	-.476	-.768	-.607
.150	.084	-.167	-.395	-.868	-.909	-.079	-.164	-.294	-.686	-.598
.250	.014	-.169	-.305	-.496	-.832	-.050	-.101	-.215	-.597	-.566
.350	-.029	-.180	-.271	-.332	-.702					
.450	-.070	-.189	-.243	-.260	-.566					
.550	-.082	-.173	-.204	-.209	-.446					
.650	-.061	-.133	-.146	-.148	-.349	-.091	-.081	-.132	-.317	-.410
.750	-.029	-.072	-.082	-.091	-.260	-.066	-.042	-.109	-.262	-.380
.850	.007	-.024	-.024	-.041	-.194	-.036	-.008	-.072	-.209	-.344
.900	.018	.003	-.006	-.020	-.158	-.020	.012	-.056	-.173	-.317
	Right side									
.025	-1.027	-.106	.393	.570	.565	-.795	-.160	.243	.353	.404
.075	-.588	-.158	.188	.353	.440	-.529	-.198	.011	.119	.220
.150	-.384	-.146	.100	.235	.343	-.332	-.173	-.077	-.025	.069
.250	-.322	-.164	.041	.148	.266					
.350	-.284	-.183	-.017	.084	.202					
.450	-.245	-.183	-.042	.050	.162					
.550	-.197	-.160	-.045	.034	.125					
.650	-.145	-.117	-.024	.032	.100	-.072	-.063	-.093	-.061	-.058
.750	-.084	-.069	.001	.039	.080	-.109	-.047	-.068	-.077	-.095
.850	-.002	.005	.045	.066	.064	-.057	.012	-.010	-.032	-.090
.900	.028	.030								

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(q) $\alpha = 15.8°$; M = 0.80

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	-.029	.863	.410	.172	-.497	.311	.776	.360	-.237	-.611
.025	-.993	.072	.343	.061	-.463	-.044	-.069	-.342	-.592	-.737
.075	-.642	-.120	.004	-.123	-.428	-.116	-.165	-.345	-.493	-.670
.150	-.314	-.214	-.205	-.254	-.415	-.104	-.177	-.317	-.430	-.588
.250	-.173	-.159	-.192	-.220	-.360	-.094	-.182	-.301	-.390	-.524
.350	-.102	-.131	-.203	-.213	-.308	-.097	-.205	-.281	-.348	-.460
.450	-.069	-.143	-.211	-.208	-.261	-.091	-.217	-.259	-.302	-.392
.550	-.048	-.135	-.183	-.173	-.214	-.069	-.180	-.211	-.243	-.325
.650	-.029	-.078	-.119	-.116	-.161	-.053	-.140	-.159	-.186	-.269
.750	-.016	-.050	-.084	-.081	-.111	-.037	-.073	-.093	-.120	-.194
.850	-.010	-.005	-.029	-.025	-.054	-.006	-.002	-.028	-.058	-.132
.900	.004	.034	-.005	.007	-.018	-.186	-.161	-.134	-.142	-.136
Right side										
.025	.354	-.754	-.971	-.483	-.307	-.465	-.288	-.054	.016	.040
.075	-.051	-.435	-.653	-.260	-.185	-.404	-.277	-.110	-.025	.082
.150	-.246	-.262	-.314	-.169	-.074	-.363	-.221	-.090	-.020	.140
.250	-.237	-.156	-.156	-.090	.032	-.328	-.170	-.063	.011	.190
.350	-.252	-.099	-.075	-.031	.096	-.301	-.155	-.044	.027	.204
.450	-.241	-.084	-.032	.008	.131	-.281	-.159	-.047	.037	.201
.550	-.206	-.062	-.011	.034	.129	-.235	-.131	-.029	.045	.183
.650	-.145	-.029	.006	.051	.116	-.176	-.084	-.011	.055	.163
.750	-.108	-.010	.020	.054	.102	-.110	-.031	.017	.061	.134
.850	-.042	.034	.045	.063	.085	-.044	.036	.045	.078	.114
.900	-.026	.049	.041	.063	.075	-.022	.048	.039	.066	.088
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.276	.671	.425	-.011	-.478	.202	.577	.117	-.351	-.665
.025	.270	-.112	-1.093	-1.272	-.822	.149	-.241	-1.177	-.783	-.556
.075	.105	-.161	-.578	-1.142	-.802	-.035	-.224	-.616	-.752	-.538
.150	.016	-.182	-.432	-.864	-.776	-.127	-.174	-.302	-.681	-.527
.250	-.034	-.187	-.330	-.530	-.719	-.076	-.088	-.224	-.593	-.503
.350	-.072	-.202	-.282	-.357	-.633					
.450	-.102	-.206	-.256	-.272	-.530					
.550	-.107	-.185	-.215	-.208	-.433					
.650	-.076	-.128	-.147	-.139	-.348	-.099	-.053	-.116	-.317	-.371
.750	-.041	-.059	-.072	-.073	-.275	-.072	-.013	-.084	-.255	-.343
.850	-.009	-.002	-.008	-.026	-.214	-.047	.033	-.052	-.198	-.308
.900	.006	.030	.023	.001	-.179	-.025	.048	-.020	-.163	-.287
Right side										
.025	-1.186	-.140	.335	.540	.628	-.934	-.170	.215	.372	.474
.075	-.598	-.180	.131	.317	.477	-.639	-.218	-.028	.127	.280
.150	-.421	-.147	.056	.214	.383	-.358	-.170	-.113	-.033	.113
.250	-.357	-.158	.000	.133	.312					
.350	-.319	-.173	-.044	.070	.248					
.450	-.272	-.162	-.063	.045	.204					
.550	-.228	-.138	-.061	.037	.161					
.650	-.164	-.088	-.032	.046	.134	-.095	-.043	-.058	-.052	-.038
.750	-.095	-.037	.003	.054	.107	-.118	-.010	-.049	-.054	-.076
.850	-.012	.046	.053	.084	.084	-.054	.056	.015	-.004	-.074
.900		.069	.068	.077	.057		.068	.024	-.004	-.091

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Concluded

(r) $\alpha = 15.9°$; M = 0.85

$\dfrac{x}{c_v}$	\multicolumn{5}{c}{C_p for – $z/b_v = 0.11$}	\multicolumn{5}{c}{C_p for – $z/b_v = 0.38$}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side
.000	.392	.914	.581	.204	-.613	.429	.776	.400	-.250	-.585
.025	-1.096	-.065	.261	-.067	-.582	.200	-.174	-.483	-.666	-.740
.075	-.538	-.258	-.045	-.225	-.525	.013	-.252	-.454	-.570	-.682
.150	-.260	-.322	-.291	-.354	-.497	-.059	-.240	-.403	-.510	-.611
.250	-.185	-.225	-.266	-.297	-.434	-.095	-.217	-.356	-.465	-.551
.350	-.149	-.162	-.245	-.274	-.366	-.132	-.232	-.330	-.409	-.484
.450	-.121	-.162	-.242	-.253	-.301	-.145	-.241	-.291	-.348	-.417
.550	-.091	-.142	-.205	-.200	-.235	-.114	-.190	-.232	-.283	-.352
.650	-.048	-.076	-.124	-.136	-.174	-.085	-.143	-.175	-.215	-.292
.750	-.020	-.045	-.092	-.094	-.123	-.046	-.078	-.104	-.138	-.218
.850	.013	.000	-.031	-.031	-.062	.008	-.007	-.032	-.067	-.154
.900	.035	.033	.000	-.003	-.025	-.162	-.162	-.138	-.141	-.136
Right side
.025	.269	-.620	-1.099	-.543	-.433	-1.102	-.285	-.052	-.047	-.035
.075	-.172	-.803	-.718	-.338	-.302	-.770	-.323	-.147	-.101	-.005
.150	-.466	-.335	-.354	-.253	-.175	-.550	-.276	-.153	-.095	.051
.250	-.333	-.241	-.229	-.175	-.057	-.326	-.224	-.127	-.061	.116
.350	-.260	-.164	-.162	-.105	.022	-.297	-.203	-.107	-.040	.139
.450	-.228	-.129	-.104	-.053	.075	-.265	-.194	-.099	-.021	.153
.550	-.177	-.091	-.059	-.014	.096	-.215	-.156	-.073	-.002	.150
.650	-.115	-.041	-.021	.011	.092	-.148	-.102	-.041	.021	.139
.750	-.063	-.014	.006	.030	.088	-.071	-.040	-.005	.040	.119
.850	.005	.031	.032	.050	.076	.004	.030	.036	.062	.099
.900	.025	.047	.040	.048	.064	.034	.041	.033	.048	.072

$\dfrac{x}{c_v}$	\multicolumn{5}{c}{$z/b_v = 0.66$}	\multicolumn{5}{c}{$z/b_v = 0.93$}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side
.000	.302	.665	.451	.102	-.316	.256	.576	.175	-.293	-.639
.025	.315	-.153	-1.209	-1.375	-.811	.160	-.268	-1.147	-.869	-.548
.075	.134	-.194	-.670	-1.192	-.794	-.039	-.268	-.696	-.827	-.544
.150	.020	-.211	-.478	-.875	-.770	-.162	-.198	-.369	-.737	-.537
.250	-.053	-.213	-.358	-.611	-.724	-.082	-.113	-.228	-.644	-.518
.350	-.105	-.224	-.300	-.439	-.649					
.450	-.142	-.224	-.272	-.315	-.551					
.550	-.141	-.198	-.226	-.236	-.460					
.650	-.103	-.139	-.151	-.158	-.374	-.095	-.057	-.120	-.331	-.386
.750	-.045	-.064	-.073	-.087	-.301	-.058	-.015	-.080	-.271	-.356
.850	.003	.002	-.002	-.029	-.237	-.019	.027	-.041	-.210	-.320
.900	.025	.030	.027	-.002	-.211	.003	.053	-.019	-.175	-.303
Right side
.025	-1.369	-.162	.292	.513	.643	-1.227	-.200	.187	.361	.486
.075	-1.059	-.214	.088	.284	.466	-.742	-.264	-.073	.105	.283
.150	-.396	-.187	.015	.182	.372	-.409	-.196	-.161	-.073	.106
.250	-.323	-.197	-.046	.098	.287					
.350	-.294	-.208	-.090	.040	.227					
.450	-.252	-.190	-.104	.010	.185					
.550	-.205	-.159	-.094	.001	.149					
.650	-.136	-.101	-.056	.020	.126	-.085	-.049	-.066	-.070	-.050
.750	-.065	-.041	-.018	.037	.099	-.083	-.021	-.056	-.073	-.089
.850	.025	.046	.050	.075	.076	-.018	.058	.016	-.013	-.079
.900	.060	.070	.064	.069	.044	.005	.070	.029	-.017	-.107

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$

(a) $\alpha = -9.4°$; $M = 0.60$

$z/b_v = 0.11$ and $z/b_v = 0.38$

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.11$ $\beta = -3.9°$	$\beta = 0°$	$\beta = 3.9°$	$\beta = 7.9°$	$\beta = 12.7°$	C_p for — $z/b_v = 0.38$ $\beta = -3.9°$	$\beta = 0°$	$\beta = 3.9°$	$\beta = 7.9°$	$\beta = 12.7°$
Left side										
.000	.728	.819	.554	-.150	-1.033	.452	.805	.540	-.109	-.723
.025	.246	-.106	-.558	-.982	-2.182	.338	-.091	-.606	-1.171	-1.074
.075	.105	-.122	-.364	-.606	-1.102	.128	-.146	-.451	-.916	-1.056
.150	-.002	-.168	-.330	-.453	-.684	.025	-.168	-.362	-.531	-.962
.250	-.039	-.137	-.237	-.294	-.403	-.057	-.197	-.326	-.392	-.835
.350	-.066	-.146	-.216	-.239	-.316	-.121	-.226	-.335	-.344	-.679
.450	-.114	-.170	-.216	-.216	-.272	-.162	-.244	-.323	-.314	-.536
.550	-.112	-.150	-.180	-.173	-.221	-.160	-.219	-.264	-.260	-.435
.650	-.085	-.110	-.132	-.114	-.182	-.135	-.177	-.203	-.207	-.369
.750	-.057	-.064	-.066	-.071	-.157	-.094	-.113	-.116	-.146	-.318
.850	.000	-.006	-.009	-.009	-.102	-.009	-.031	-.030	-.071	-.249
.900	.027	.029	.041	.025	-.061	.025	.011	.023	-.023	-.201
Right side										
.025	-.484	-.064	.283	.545	.781	-.797	-.106	.349	.595	.762
.075	-.418	-.155	.080	.280	.491	-.475	-.168	.125	.328	.505
.150	-.345	-.170	.000	.148	.314	-.379	-.184	.023	.171	.328
.250	-.260	-.146	-.018	.075	.195	-.340	-.204	-.059	.057	.183
.350	-.240	-.161	-.080	-.004	.096	-.336	-.230	-.114	-.025	.089
.450	-.228	-.181	-.118	-.061	.015	-.327	-.244	-.153	-.093	-.005
.550	-.185	-.155	-.109	-.073	-.017	-.279	-.221	-.153	-.112	-.054
.650	-.133	-.106	-.082	-.061	-.024	-.203	-.179	-.130	-.102	-.067
.750	-.073	-.062	-.045	-.041	-.021	-.117	-.099	-.080	-.066	-.056
.850	.011	.005	.018	.016	.011	-.009	-.011	.025	.005	-.024
.900	.039	.038	.046	.037	.015	.018	.007	.032	.005	-.049

$z/b_v = 0.66$ and $z/b_v = 0.93$

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.66$ $\beta = -3.9°$	$\beta = 0°$	$\beta = 3.9°$	$\beta = 7.9°$	$\beta = 12.7°$	C_p for — $z/b_v = 0.93$ $\beta = -3.9°$	$\beta = 0°$	$\beta = 3.9°$	$\beta = 7.9°$	$\beta = 12.7°$
Left side										
.000	.397	.812	.499	-.116	-.352	.498	.734	.194	-.153	-.375
.025	.363	-.150	-.872	-1.403	-.792	.308	-.496	-1.879	-1.360	-.467
.075	.146	-.204	-.576	-1.066	-.824	-.005	-.543	-1.203	-.451	-.454
.150	.004	-.257	-.512	-.619	-.817	-.228	-.567	-1.150	-.440	-.449
.250	-.105	-.299	-.474	-.440	-.674	-.320	-.534	-.790	-.437	-.463
.350	-.185	-.328	-.471	-.380	-.610					
.450	-.224	-.352	-.440	-.360	-.536					
.550	-.238	-.317	-.371	-.344	-.511					
.650	-.192	-.232	-.271	-.317	-.511	-.235	-.279	-.262	-.444	-.490
.750	-.130	-.153	-.164	-.305	-.502	-.126	-.150	-.157	-.428	-.461
.850	-.041	-.068	-.068	-.276	-.456	-.030	-.040	-.066	-.394	-.412
.900	-.018	-.033	-.023	-.250	-.431	.014	.007	-.020	-.362	-.392
Right side										
.025	-.831	-.146	.387	.633	.774	-1.416	-.439	.330	.574	.783
.075	-.619	-.226	.121	.344	.510	-1.381	-.545	-.027	.201	.431
.150	-.498	-.250	.000	.173	.330	-1.276	-.547	-.223	-.041	.158
.250	-.491	-.308	-.102	.027	.174					
.350	-.502	-.348	-.180	-.084	.050					
.450	-.448	-.354	-.228	-.157	-.054					
.550	-.384	-.323	-.228	-.189	-.111					
.650	-.281	-.235	-.182	-.178	-.129	-.231	-.230	-.189	-.207	-.196
.750	-.164	-.153	-.098	-.134	-.127	-.139	-.144	-.112	-.191	-.194
.850	-.046	-.040	-.004	-.071	-.111	-.018	-.028	-.002	-.130	-.159
.900	.000	-.006	.023	-.068	-.136	.027	.023	.023	-.143	-.182

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(b) $\alpha = -9.6°$; M = 0.80

$\frac{x}{c_v}$	C_p for $z/b_v = 0.11$					C_p for $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.849	.883	.720	.367	-.102	.663	.869	.732	.375	-.098
.025	.232	-.098	-.549	-.894	-1.422	.319	-.074	-.988	-1.622	-.976
.075	.095	-.128	-.387	-.797	-1.118	.122	-.151	-.443	-1.406	-.965
.150	-.015	-.187	-.356	-.503	-.881	.012	-.178	-.364	-.384	-.833
.250	-.052	-.158	-.248	-.281	-.601	-.080	-.217	-.330	-.299	-.765
.350	-.095	-.167	-.231	-.229	-.389	-.172	-.273	-.347	-.310	-.595
.450	-.162	-.215	-.240	-.215	-.290	-.248	-.341	-.350	-.296	-.439
.550	-.174	-.197	-.202	-.177	-.229	-.269	-.301	-.289	-.243	-.357
.650	-.151	-.154	-.145	-.125	-.198	-.230	-.223	-.221	-.201	-.320
.750	-.101	-.083	-.073	-.073	-.169	-.138	-.126	-.132	-.148	-.302
.850	-.026	-.005	-.001	-.012	-.113	-.042	-.018	-.037	-.076	-.267
.900	.015	.037	.043	.021	-.073	.009	.032	.018	-.035	-.232
Right side										
.025	-.422	-.028	.286	.541	.773	-.908	.078	.341	.587	.767
.075	-.417	-.147	.082	.283	.491	-.462	-.157	.117	.320	.512
.150	-.360	-.178	-.011	.142	.314	-.376	-.184	.009	.172	.337
.250	-.266	-.152	-.043	.070	.198	-.328	-.221	-.081	.052	.193
.350	-.246	-.186	-.111	-.020	.085	-.337	-.272	-.167	-.052	.076
.450	-.254	-.220	-.173	-.098	-.014	-.342	-.318	-.244	-.147	-.046
.550	-.200	-.195	-.178	-.127	-.063	-.293	-.301	-.262	-.192	-.115
.650	-.136	-.141	-.146	-.118	-.081	-.212	.227	-.231	-.186	-.137
.750	-.071	-.078	-.099	-.092	-.084	-.118	.098	-.134	-.127	-.124
.850	.013	.014	-.008	-.020	-.044	-.018	.006	-.015	-.049	-.092
.900	.047	.040	.024	.003	-.032	.021	.029	.000	-.038	-.113

$\frac{x}{c_v}$	C_p for $z/b_v = 0.66$					C_p for $z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.666	.896	.743	.392	.007	.755	.840	.677	.311	.061
.025	.346	-.108	-1.117	-1.627	-.845	.225	.402	-1.181	-1.209	-.403
.075	.134	-.186	-.487	-1.383	-.860	-.079	.621	-.601	-.349	-.404
.150	-.017	-.258	-.472	-.714	-.793	-.339	.657	-.548	-.343	-.398
.250	-.150	-.346	-.425	-.467	-.538	-.464	.728	-.528	-.348	-.400
.350	-.283	-.445	-.437	-.333	-.502					
.450	-.399	-.515	-.432	-.285	-.456					
.550	-.435	-.447	-.371	-.282	-.467					
.650	-.337	-.260	-.291	-.294	-.480	-.339	.255	-.394	-.372	-.427
.750	-.186	-.155	-.218	-.316	-.488	-.213	.117	-.368	-.383	-.416
.850	-.101	-.043	-.140	-.330	-.467	-.130	.006	-.307	-.366	-.384
.900	-.071	.002	-.108	-.325	-.459	-.110	.035	-.271	-.354	-.368
Right side										
.025	-1.097	-.095	.379	.625	.786	-.875	.370	.260	.585	.799
.075	-.481	-.204	.122	.344	.533	-.610	.664	-.100	.201	.453
.150	-.444	-.244	-.006	.178	.353	-.577	.653	-.339	-.073	.164
.250	-.434	-.349	-.155	.013	.183					
.350	-.459	-.445	-.288	-.133	.021					
.450	-.434	-.502	-.388	-.259	-.112					
.550	-.360	-.444	-.426	-.331	-.206					
.650	-.287	-.260	-.345	-.308	-.230	-.360	.232	-.279	-.301	-.272
.750	-.212	-.146	-.178	-.214	-.211	-.354	.104	-.187	-.243	-.247
.850	-.126	-.008	-.058	-.124	-.157	-.295	.009	-.091	-.157	-.169
.900	-.083	.038	-.030	-.128	-.188	-.254	.052	-.076	-.165	-.188

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(c) $\alpha = -9.7°$; $M = 0.85$

$z/b_v = 0.11$ and $z/b_v = 0.38$

$\frac{x}{c_v}$	C_p for $z/b_v = 0.11$					C_p for $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.880	.907	.758	.468	.051	.702	.885	.771	.487	.042
.025	.250	-.105	-.549	-.870	-1.382	.326	-.078	-1.012	-1.417	-.999
.075	.109	-.136	-.426	-.794	-1.083	.130	-.163	-.396	-1.278	-.998
.150	-.016	-.210	-.383	-.656	-.834	.019	-.197	-.376	-1.127	-.825
.250	-.048	-.177	-.263	-.303	-.627	-.078	-.230	-.341	-.227	-.776
.350	-.093	-.192	-.245	-.216	-.424	-.171	-.296	-.355	-.233	-.602
.450	-.174	-.252	-.255	-.200	-.304	-.273	-.404	-.366	-.268	-.439
.550	-.206	-.262	-.212	-.169	-.233	-.320	-.431	-.305	-.233	-.357
.650	-.197	-.223	-.155	-.127	-.202	-.321	-.350	-.232	-.206	-.318
.750	-.143	-.129	-.086	-.080	-.179	-.203	-.161	-.146	-.160	-.306
.850	-.047	-.028	-.004	-.025	-.135	-.061	-.030	-.050	-.093	-.281
.900	.006	.025	.033	.014	-.096	-.006	.029	.009	-.049	-.247
Right side										
.025	-.414	-.030	.300	.542	.767	-.948	-.081	.348	.581	.760
.075	-.444	-.163	.086	.283	.487	-.527	-.172	.119	.318	.508
.150	-.387	-.199	-.017	.143	.308	-.374	-.197	.007	.170	.336
.250	-.267	-.173	-.047	.066	.192	-.327	-.227	-.082	.049	.186
.350	-.254	-.211	-.119	-.023	.081	-.333	-.298	-.176	-.060	.066
.450	-.253	-.265	-.193	-.117	-.031	-.343	-.380	-.275	-.170	-.062
.550	-.201	-.267	-.219	-.156	-.093	-.298	-.415	-.326	-.240	-.153
.650	-.141	-.206	-.195	-.166	-.122	-.216	-.350	-.319	-.267	-.197
.750	-.076	-.120	-.143	-.145	-.137	-.124	-.110	-.213	-.206	-.189
.850	.010	-.008	-.034	-.055	-.088	-.028	-.002	-.039	-.092	-.137
.900	.049	.030	.007	-.021	-.072	.014	.025	-.023	-.065	-.150

$z/b_v = 0.66$ and $z/b_v = 0.93$

$\frac{x}{c_v}$	C_p for $z/b_v = 0.66$					C_p for $z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.704	.909	.781	.495	.102	.802	.879	.731	.459	.173
.025	.360	-.093	-1.107	-1.485	-.887	.249	-.312	-1.027	-1.109	-.387
.075	.154	-.176	-.804	-1.308	-.853	-.046	-.518	-.496	-.340	-.381
.150	-.003	-.245	-.461	-1.130	-.766	-.310	-.714	-.455	-.338	-.381
.250	-.138	-.330	-.403	-.394	-.566	-.456	-.764	-.441	-.343	-.387
.350	-.276	-.462	-.414	-.278	-.516					
.450	-.416	-.595	-.428	-.280	-.435					
.550	-.514	-.685	-.378	-.297	-.434					
.650	-.557	-.569	-.302	-.314	-.449	-.628	-.395	-.398	-.367	-.409
.750	-.233	-.122	-.246	-.328	-.462	-.303	-.303	-.385	-.371	-.409
.850	-.106	-.054	-.189	-.348	-.455	-.184	-.149	-.339	-.360	-.390
.900	-.086	-.009	-.162	-.357	-.459	-.153	-.086	-.316	-.354	-.380
Right side										
.025	-1.071	-.084	.386	.622	.793	-.748	-.293	.287	.598	.818
.075	-.774	-.192	.133	.343	.535	-.468	-.607	-.076	.219	.464
.150	-.416	-.234	.007	.182	.358	-.457	-.634	-.329	-.072	.170
.250	-.387	-.339	-.146	.016	.185					
.350	-.417	-.472	-.282	-.140	.016					
.450	-.417	-.589	-.431	-.283	-.139					
.550	-.354	-.660	-.524	-.397	-.267					
.650	-.287	-.507	-.548	-.474	-.347	-.368	-.368	-.521	-.649	-.525
.750	-.240	-.125	-.272	-.432	-.334	-.368	-.278	-.345	-.317	-.276
.850	-.176	-.022	-.083	-.143	-.189	-.327	-.148	-.183	-.174	-.174
.900	-.141	.022	-.063	-.142	-.204	-.290	-.078	-.146	-.166	-.183

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(d) α = -9.8°; M = 0.90

$\frac{x}{c_v}$	C_p for – $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for – $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
		z/b_v = 0.11					z/b_v = 0.38			
					Left side					
.000	.913	.926	.817	.568	.162	.751	.893	.821	.587	.152
.025	.238	-.095	-.470	-.822	-1.239	.320	-.069	-.893	-1.228	-.960
.075	.104	-.133	-.450	-.740	-.972	.126	-.162	-.681	-1.131	-.948
.150	-.020	-.211	-.417	-.675	-.796	.017	-.191	-.443	-1.057	-.750
.250	-.054	-.178	-.279	-.620	-.683	-.070	-.218	-.295	-.867	-.745
.350	-.096	-.183	-.235	-.274	-.510	-.170	-.285	-.362	-.190	-.649
.450	-.183	-.250	-.257	-.172	-.378	-.295	-.381	-.335	-.147	-.488
.550	-.238	-.295	-.215	-.137	-.286	-.353	-.451	-.317	-.180	-.386
.650	-.272	-.295	-.165	-.112	-.233	-.427	-.506	-.237	-.195	-.333
.750	-.280	-.189	-.093	-.084	-.208	-.449	-.266	-.157	-.179	-.305
.850	-.127	-.033	-.013	-.039	-.161	-.229	.021	-.063	-.124	-.282
.900	-.047	.022	.029	-.004	-.135	-.079	.038	-.007	-.083	-.257
					Right side					
.025	-.364	-.023	.294	.535	.787	-.865	-.089	.338	.571	.772
.075	-.479	-.167	.079	.284	.508	-.696	-.183	.115	.313	.522
.150	-.444	-.209	-.021	.139	.330	-.529	-.198	.011	.169	.351
.250	-.312	-.173	-.042	.070	.217	-.293	-.221	-.077	.051	.206
.350	-.258	-.211	-.117	-.031	.094	-.348	-.291	-.171	-.063	.086
.450	-.266	-.259	-.194	-.125	-.022	-.312	-.369	-.277	-.178	-.048
.550	-.213	-.303	-.245	-.183	-.096	-.329	-.453	-.351	-.262	-.152
.650	-.158	-.282	-.259	-.216	-.148	-.236	-.486	-.406	-.325	-.227
.750	-.090	-.191	-.297	-.270	-.219	-.142	-.339	-.431	-.369	-.288
.850	-.003	-.026	-.119	-.156	-.192	-.052	.021	-.247	-.286	-.282
.900	.033	.025	-.049	-.095	-.189	-.016	.033	-.073	-.203	-.340

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
		z/b_v = 0.66					z/b_v = 0.93			
					Left side					
.000	.748	.925	.831	.587	.202	.838	.902	.791	.548	.217
.025	.368	-.069	-.977	-1.291	-.836	.296	-.206	-.881	-1.041	-.377
.075	.165	-.154	-.823	-1.156	-.815	-.007	-.439	-.415	-.341	-.380
.150	.017	-.219	-.689	-1.075	-.759	-.268	-.643	-.366	-.333	-.376
.250	-.114	-.298	-.357	-.658	-.619	-.434	-.734	-.349	-.335	-.377
.350	-.256	-.417	-.279	-.276	-.578					
.450	-.407	-.557	-.374	-.222	-.422					
.550	-.529	-.673	-.383	-.281	-.408					
.650	-.604	-.687	-.290	-.313	-.420	-.857	-.378	-.353	-.361	-.398
.750	-.671	-.293	-.270	-.346	-.434	-.735	-.390	-.361	-.363	-.397
.850	-.561	-.038	-.254	-.371	-.438	-.396	-.342	-.347	-.355	-.389
.900	-.284	-.018	-.246	-.385	-.449	-.341	-.298	-.338	-.354	-.382
					Right side					
.025	-.967	-.070	.389	.619	.799	-.714	-.230	.317	.615	.847
.075	-.825	-.185	.143	.351	.548	-.399	-.535	-.041	.240	.498
.150	-.660	-.222	.023	.194	.381	-.375	-.582	-.299	-.043	.208
.250	-.341	-.309	-.117	.034	.212					
.350	-.296	-.438	-.262	-.120	.048					
.450	-.380	-.566	-.395	-.276	-.111					
.550	-.374	-.653	-.538	-.387	-.247					
.650	-.299	-.690	-.595	-.492	-.346	-.371	-.439	-.799	-.731	-.570
.750	-.287	-.347	-.654	-.561	-.462	-.372	-.406	-.681	-.811	-.655
.850	-.260	-.013	-.607	-.589	-.478	-.349	-.318	-.435	-.617	-.613
.900	-.242	.014	-.291	-.575	-.494	-.333	-.258	-.358	-.387	-.412

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(e) $\alpha = -9.8°$; $M = 0.92$

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.927	.940	.831	.573	.191	.746	.898	.829	.593	.185
.025	.270	-.095	-.468	-.876	-1.215	.341	-.066	-.866	-1.184	-.797
.075	.126	-.135	-.490	-.749	-.969	.142	-.172	-.728	-1.094	-.798
.150	-.006	-.222	-.480	-.715	-.830	.031	-.199	-.578	-1.052	-.806
.250	-.043	-.188	-.513	-.722	-.661	-.058	-.218	-.536	-.965	-.772
.350	-.090	-.192	-.291	-.479	-.545	-.160	-.285	-.388	-.653	-.642
.450	-.177	-.256	-.279	-.258	-.439	-.273	-.372	-.365	-.149	-.510
.550	-.252	-.315	-.236	-.120	-.353	-.355	-.460	-.258	-.112	-.442
.650	-.279	-.334	-.186	-.105	-.307	-.429	-.524	-.283	-.161	-.417
.750	-.338	-.332	-.130	-.094	-.282	-.459	-.535	-.212	-.191	-.410
.850	-.246	-.075	-.046	-.067	-.245	-.450	-.067	-.123	-.171	-.403
.900	-.116	.006	-.004	-.040	-.212	-.271	.031	-.060	-.140	-.361
Right side										
.025	-.379	-.028	.292	.555	.795	-.861	-.096	.329	.579	.769
.075	-.528	-.177	.076	.299	.506	-.768	-.197	.109	.322	.516
.150	-.490	-.232	-.028	.152	.333	-.619	-.211	.001	.178	.348
.250	-.470	-.180	-.051	.072	.211	-.495	-.231	-.081	.058	.207
.350	-.293	-.216	-.126	-.027	.090	-.343	-.298	-.176	-.055	.080
.450	-.273	-.274	-.207	-.126	-.026	-.270	-.368	-.277	-.177	-.054
.550	-.206	-.324	-.278	-.194	-.106	-.288	-.463	-.375	-.258	-.157
.650	-.175	-.311	-.279	-.226	-.164	-.276	-.519	-.420	-.335	-.236
.750	-.118	-.344	-.362	-.322	-.258	-.178	-.489	-.456	-.377	-.309
.850	-.025	-.067	-.281	-.263	-.235	-.087	-.043	-.471	-.386	-.309
.900	.010	.006	-.177	-.242	-.309	-.054	.051	-.442	-.412	-.385
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.749	.930	.844	.600	.228	.845	.916	.810	.568	.245
.025	.385	-.058	-.941	-1.230	-.866	.327	-.211	-.891	-1.027	-.458
.075	.183	-.154	-.811	-1.114	-.865	.026	-.413	-.416	-.374	-.435
.150	.034	-.223	-.764	-1.074	-.789	-.238	-.631	-.372	-.364	-.418
.250	-.098	-.292	-.675	-.970	-.579	-.411	-.737	-.363	-.368	-.418
.350	-.235	-.408	-.321	-.428	-.471					
.450	-.384	-.545	-.240	-.279	-.451					
.550	-.514	-.667	-.340	-.294	-.462					
.650	-.590	-.730	-.340	-.349	-.476	-.829	-.395	-.374	-.382	-.431
.750	-.656	-.644	-.341	-.382	-.490	-.913	-.408	-.389	-.386	-.431
.850	-.648	-.126	-.350	-.403	-.480	-.684	-.404	-.382	-.383	-.421
.900	-.671	-.050	-.353	-.420	-.485	-.527	-.379	-.376	-.385	-.413
Right side										
.025	-.953	-.072	.388	.624	.799	-.711	-.214	.330	.631	.850
.075	-.832	-.186	.143	.360	.549	-.394	-.518	-.030	.262	.503
.150	-.748	-.224	.023	.204	.373	-.375	-.589	-.275	-.021	.208
.250	-.681	-.313	-.118	.046	.212					
.350	-.151	-.434	-.260	-.114	.047					
.450	-.272	-.566	-.396	-.262	-.110					
.550	-.358	-.666	-.539	-.379	-.243					
.650	-.339	-.742	-.606	-.475	-.362	-.390	-.484	-.834	-.717	-.574
.750	-.343	-.665	-.665	-.557	-.458	-.391	-.456	-.877	-.797	-.655
.850	-.337	-.133	-.663	-.581	-.497	-.375	-.400	-.674	-.788	-.658
.900	-.326	-.037	-.691	-.612	-.530	-.359	-.351	-.564	-.699	-.619

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(f) $\alpha = 0°$; $M = 0.60$

$\dfrac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.606	.675	.344	-.455	-1.251	.271	.749	.314	-.393	-1.072
.025	.287	-.090	-.546	-1.109	-1.849	.354	-.086	-.649	-1.040	-1.233
.075	.142	-.111	-.378	-.665	-1.435	.170	-.125	-.447	-.920	-.920
.150	.050	-.136	-.318	-.464	-.874	.075	-.136	-.341	-.674	-.893
.250	.029	-.100	-.212	-.284	-.444	.006	-.146	-.295	-.409	-.817
.350	-.017	-.113	-.189	-.231	-.336	-.052	-.178	-.290	-.319	-.707
.450	-.061	-.125	-.178	-.195	-.295	-.089	-.194	-.270	-.266	-.587
.550	-.052	-.109	-.141	-.151	-.293	-.089	-.159	-.208	-.220	-.470
.650	-.038	-.077	-.090	-.109	-.217	-.066	-.123	-.155	-.176	-.392
.750	-.015	-.026	-.031	-.059	-.148	-.029	-.061	-.079	-.123	-.313
.850	.029	.025	.025	-.010	-.072	.024	.013	.006	-.052	-.233
.900	.052	.059	.057	.020	-.028	.059	.052	.048	-.008	-.182
Right side										
.025	-.482	-.035	.321	.612	.831	-.648	-.061	.390	.610	.744
.075	-.394	-.118	.130	.354	.556	-.447	-.116	.169	.382	.539
.150	-.315	-.129	.050	.218	.381	-.350	-.132	.075	.239	.390
.250	-.228	-.104	.020	.156	.275	-.292	-.146	.013	.137	.268
.350	-.202	-.118	-.024	.075	.174	-.274	-.159	-.035	.068	.183
.450	-.184	-.136	-.067	.015	.098	-.200	-.136	-.084	-.022	.049
.550	-.140	-.104	-.054	.008	.073	-.211	-.150	-.077	-.015	.052
.650	-.080	-.070	-.033	.006	.061	-.147	-.111	-.054	-.015	.034
.750	-.031	-.021	-.001	.024	.048	-.070	-.040	-.005	.004	.031
.850	.043	.041	.050	.057	.073	.024	.036	.059	.047	.031
.900	.066	.066	.064	.066	.068	.052	.052	.059	.047	.011
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.163	.751	.220	-.324	-.654	.394	.779	.036	-.248	-.509
.025	.403	-.109	-.959	-.987	-.720	.495	-.313	-1.357	-.701	-.484
.075	.209	-.139	-.546	-.936	-.743	.221	-.316	-1.152	-.367	
.150	.091	-.164	-.433	-.840	-.775	.003	-.334	-1.033	-.354	
.250	.010	-.182	-.378	-.598	-.707	-.103	-.329	-.605	-.356	-.449
.350	-.054	-.215	-.364	-.377	-.546					
.450	-.105	-.233	-.336	-.275	-.511					
.550	-.119	-.212	-.281	-.261	-.504					
.650	-.089	-.159	-.198	-.277	-.493	-.131	.180	-.185	-.397	-.438
.750	-.038	-.077	-.111	-.280	-.461	-.054	.079	-.093	-.390	-.401
.850	.017	-.003	-.024	-.252	-.408	.022	.025	-.003	-.351	-.348
.900	.031	.027	.015	-.236	-.385	.052	.064	.029	-.328	-.320
Right side										
.025	-.805	-.051	.441	.640	.742	-1.264	-.164	.542	.746	.910
.075	-.532	-.129	.208	.409	.558	-.897	-.281	.215	.421	.631
.150	-.396	-.136	.103	.271	.418	-1.038	-.297	.022	.199	.390
.250	-.375	-.178	.020	.163	.298					
.350	-.371	-.208	-.056	.057	.192					
.450	-.334	-.219	-.097	-.015	.100					
.550	-.278	-.205	-.111	-.054	.029					
.650	-.195	-.150	-.088	-.056	-.001	-.114	.129	-.111	-.112	-.086
.750	-.103	-.074	-.038	-.047	-.024	-.077	.072	-.056	-.130	-.111
.850	.006	.027	.041	-.017	-.042	.017	.041	.031	-.105	-.109
.900	.047	.050	.061	-.022	-.081	.047	.080	.061	-.118	-.136

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(g) $\alpha = 0°$; M = 0.80

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.726	.680	.546	.190	-.264	.496	.781	.549	.075	-.348
.025	.295	-.059	-.525	-1.131	-.945	.353	-.068	-.932	-1.224	-.670
.075	.158	-.092	-.367	-.851	-.840	.175	-.121	-.442	-1.046	-.627
.150	.058	-.132	-.324	-.471	-.746	.082	-.140	-.336	-.851	-.653
.250	.035	-.098	-.212	-.282	-.620	.005	-.156	-.286	-.518	-.653
.350	-.018	-.123	-.196	-.229	-.519	-.061	-.202	-.289	-.312	-.591
.450	-.070	-.149	-.188	-.197	-.463	-.122	-.231	-.283	-.236	-.513
.550	-.068	-.123	-.145	-.147	-.372	-.117	-.194	-.225	-.188	-.430
.650	-.059	-.091	-.094	-.106	-.290	-.094	-.146	-.159	-.159	-.352
.750	-.024	-.033	-.030	-.063	-.206	-.044	-.061	-.071	-.116	-.279
.850	.027	.034	.036	-.002	-.121	.026	.028	.016	-.051	-.228
.900	.061	.070	.076	.031	-.073	.067	.073	.062	-.004	-.199
Right side										
.025	-.411	.006	.340	.606	.829	-.976	-.042	.386	.617	.754
.075	-.390	-.095	.149	.360	.561	-.429	-.112	.177	.384	.549
.150	-.314	-.121	.060	.228	.402	-.335	-.123	.085	.249	.403
.250	-.216	-.098	.027	.158	.295	-.277	-.135	.014	.148	.286
.350	-.199	-.127	-.026	.075	.192	-.269	-.171	-.048	.067	.195
.450	-.186	-.146	-.073	.008	.104	-.219	-.168	-.094	-.016	.084
.550	-.137	-.120	-.068	-.009	.067	-.222	-.185	-.117	-.043	.047
.650	-.075	-.080	-.050	-.010	.043	-.146	-.124	-.088	-.043	.017
.750	-.021	-.024	-.018	.001	.028	-.053	-.045	-.032	-.021	.003
.850	.053	.047	.042	.045	.043	.040	.049	.045	.026	.000
.900	.090	.076	.060	.057	.031	.079	.079	.063	.026	-.033
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.452	.781	.497	.125	-.380	.677	.827	.537	.148	-.202
.025	.400	-.086	-1.070	-.714	-.735	.469	-.298	-.944	-.567	-.381
.075	.218	-.127	-.519	-.701	-.746	.192	-.383	-.607	-.291	-.372
.150	.104	-.165	-.393	-.664	-.747	-.038	-.393	-.577	-.288	-.366
.250	.008	-.206	-.341	-.562	-.738	-.148	-.375	-.560	-.292	-.381
.350	-.078	-.261	-.349	-.418	-.491					
.450	-.155	-.299	-.346	-.303	-.434					
.550	-.178	-.272	-.295	-.254	-.437					
.650	-.145	-.190	-.211	-.250	-.454	-.190	-.171	-.281	-.342	-.412
.750	-.068	-.071	-.125	-.260	-.456	-.090	-.036	-.232	-.347	-.393
.850	.002	.012	-.041	-.250	-.430	-.020	.067	-.169	-.333	-.345
.900	.026	.049	-.003	-.238	-.415	.003	.105	-.133	-.315	-.323
Right side										
.025	-1.055	-.030	.437	.641	.762	-1.073	-.150	.510	.770	.950
.075	-.487	-.114	.216	.419	.585	-.524	-.339	.178	.442	.663
.150	-.358	-.130	.112	.290	.450	-.577	-.343	-.015	.202	.420
.250	-.347	-.190	.011	.170	.326					
.350	-.361	-.243	-.073	.057	.204					
.450	-.345	-.279	-.143	-.030	.095					
.550	-.283	-.257	-.171	-.091	.017					
.650	-.193	-.174	-.140	-.109	-.033	-.050	-.091	-.122	-.141	-.123
.750	-.103	-.071	-.075	-.091	-.068	-.193	-.030	-.102	-.165	-.149
.850	.000	.049	.022	-.046	-.068	-.135	.088	-.018	-.127	-.133
.900	.046	.088	.051	-.053	-.108	-.091	.123	-.003	-.138	-.155

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TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(h) $\alpha = 0°$; M = 0.85

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	— $z/b_v = 0.11$					$z/b_v = 0.38$				
					Left side					
.000	.748	.685	.596	.304	-.114	.548	.795	.609	.190	-.225
.025	.291	-.052	-.515	-1.094	-.928	.347	-.064	-1.079	-1.548	-.582
.075	.154	-.090	-.374	-.862	-.776	.169	-.122	-.408	-1.364	-.588
.150	.052	-.135	-.334	-.617	-.698	.072	-.142	-.339	-.666	-.614
.250	.034	-.106	-.218	-.270	-.581	-.002	-.159	-.288	-.306	-.624
.350	-.025	-.133	-.195	-.221	-.509	-.074	-.215	-.288	-.293	-.581
.450	-.084	-.167	-.195	-.202	-.473	-.140	-.260	-.283	-.258	-.517
.550	-.086	-.143	-.151	-.149	-.404	-.147	-.230	-.226	-.201	-.445
.650	-.081	-.110	-.098	-.112	-.330	-.129	-.172	-.162	-.162	-.379
.750	-.043	-.046	-.029	-.070	-.253	-.070	-.073	-.083	-.117	-.311
.850	.015	.033	.036	-.012	-.168	.012	.030	.007	-.051	-.253
.900	.045	.073	.071	.019	-.117	.051	.076	.059	-.014	-.220
					Right side					
.025	-.387	.011	.336	.603	.837	-.982	-.042	.380	.605	.766
.075	-.393	-.099	.143	.354	.572	-.429	-.116	.170	.370	.557
.150	-.323	-.129	.059	.225	.408	-.334	-.133	.074	.240	.408
.250	-.224	-.109	.022	.151	.302	-.275	-.150	.002	.141	.290
.350	-.202	-.137	-.040	.065	.192	-.270	-.192	-.064	.058	.196
.450	-.193	-.166	-.096	-.012	.096	-.241	-.208	-.115	-.035	.084
.550	-.142	-.140	-.095	-.030	.062	-.224	-.216	-.147	-.073	.033
.650	-.083	-.097	-.075	-.034	.031	-.149	-.157	-.121	-.077	-.006
.750	-.020	-.037	-.040	-.033	.010	-.060	-.054	-.057	-.053	-.020
.850	.051	.046	.033	.024	.025	.034	.050	.028	-.001	-.024
.900	.082	.077	.051	.034	.010	.069	.081	.052	.003	-.065
	$z/b_v = 0.66$					$z/b_v = 0.93$				
					Left side					
.000	.505	.795	.563	.144	-.267	.730	.855	.619	.209	-.146
.025	.396	-.076	-1.155	-.962	-.696	.442	-.272	-.889	-.522	-.404
.075	.215	-.125	-.464	-.910	-.703	.177	-.425	-.493	-.293	-.384
.150	.095	-.165	-.374	-.770	-.706	-.060	-.429	-.437	-.293	-.368
.250	-.004	-.215	-.313	-.550	-.727	-.175	-.389	-.429	-.298	-.377
.350	-.101	-.290	-.323	-.364	-.502					
.450	-.188	-.348	-.326	-.267	-.435					
.550	-.232	-.345	-.283	-.242	-.432					
.650	-.195	-.230	-.220	-.251	-.449	-.235	-.155	-.302	-.336	-.405
.750	-.107	-.073	-.147	-.271	-.450	-.127	-.024	-.283	-.347	-.386
.850	-.022	.024	-.080	-.274	-.432	-.057	.073	-.237	-.339	-.355
.900	.001	.061	-.043	-.264	-.423	-.035	.113	-.197	-.330	-.337
					Right side					
.025	-1.145	-.030	.433	.644	.771	-1.012	-.143	.501	.779	.962
.075	-.436	-.112	.212	.419	.588	-.413	-.395	.161	.443	.673
.150	-.334	-.133	.110	.288	.457	-.435	-.382	-.039	.196	.420
.250	-.317	-.203	.005	.164	.332					
.350	-.337	-.272	-.096	.045	.207					
.450	-.328	-.326	-.180	-.060	.088					
.550	-.278	-.318	-.227	-.139	-.010					
.650	-.206	-.200	-.193	-.155	-.065	-.168	-.139	-.157	-.183	-.154
.750	-.130	-.076	-.112	-.130	-.102	-.255	-.022	-.135	-.205	-.181
.850	-.038	.056	-.006	-.073	-.091	-.208	.090	-.054	-.149	-.151
.900	.006	.097	.025	-.081	-.127	-.169	.129	-.044	-.165	-.175

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(i) α = 0°; M = 0.90

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.11$					C_p for — $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side					Left side				
.000	.780	.696	.649	.412	.032	.607	.810	.669	.325	-.110
.025	.300	-.042	-.458	-.993	-.922	.348	-.062	-.979	-1.344	-.541
.075	.165	-.079	-.371	-.778	-.739	.173	-.125	-.436	-1.216	-.542
.150	.061	-.140	-.341	-.648	-.647	.079	-.137	-.319	-1.092	-.556
.250	.036	-.105	-.213	-.396	-.526	.005	-.156	-.272	-.236	-.566
.350	-.027	-.136	-.193	-.201	-.470	-.077	-.227	-.274	-.245	-.553
.450	-.091	-.182	-.195	-.182	-.459	-.160	-.291	-.281	-.270	-.507
.550	-.101	-.171	-.146	-.146	-.416	-.183	-.295	-.224	-.198	-.459
.650	-.114	-.141	-.094	-.111	-.363	-.192	-.227	-.158	-.157	-.414
.750	-.075	-.063	-.024	-.067	-.305	-.124	-.087	-.078	-.112	-.360
.850	.001	.031	.047	-.015	-.234	-.007	.035	.011	-.052	-.319
.900	.043	.081	.079	.031	-.190	.047	.090	.064	-.011	-.283
	Right side					Right side				
.025	-.331	.031	.348	.616	.846	-.921	-.028	.377	.610	.771
.075	-.398	-.086	.161	.371	.580	-.554	-.110	.181	.383	.566
.150	-.344	-.120	.068	.241	.416	-.327	-.126	.080	.250	.414
.250	-.219	-.106	.032	.163	.307	-.260	-.151	.008	.151	.294
.350	-.200	-.139	-.032	.074	.194	-.258	-.200	-.062	.066	.199
.450	-.195	-.182	-.099	-.011	.094	-.239	-.235	-.134	-.032	.082
.550	-.141	-.165	-.111	-.042	.045	-.219	-.268	-.181	-.086	.016
.650	-.082	-.124	-.100	-.056	.004	-.144	-.207	-.179	-.112	-.035
.750	-.020	-.051	-.071	-.063	-.031	-.055	-.066	-.110	-.102	-.064
.850	.060	.043	.013	-.003	-.017	.035	.057	.007	-.038	-.070
.900	.087	.081	.043	.016	-.044	.071	.086	.035	-.020	-.110

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side					Left side				
.000	.567	.812	.629	.261	-.117	.781	.884	.687	.313	-.171
.025	.399	-.066	-1.067	-.874	-.594	.458	-.215	-.895	-.493	-.609
.075	.222	-.114	-.855	-.821	-.598	.186	-.427	-.400	-.288	-.394
.150	.103	-.157	-.305	-.750	-.609	-.059	-.484	-.328	-.283	-.371
.250	.001	-.210	-.245	-.588	-.605	-.157	-.448	-.321	-.281	-.374
.350	-.104	-.322	-.278	-.422	-.580					
.450	-.215	-.416	-.305	-.305	-.544					
.550	-.303	-.432	-.268	-.222	-.454					
.650	-.326	-.371	-.212	-.226	-.426	-.539	-.183	-.287	-.308	-.410
.750	-.205	-.054	-.155	-.260	-.424	-.142	-.058	-.283	-.321	-.404
.850	-.037	.040	-.104	-.281	-.415	-.066	.039	-.250	-.313	-.386
.900	-.010	.078	-.072	-.284	-.412	-.054	.081	-.229	-.311	-.370
	Right side					Right side				
.025	-1.047	-.009	.434	.651	.782	-.953	-.093	.505	.796	.977
.075	-.787	-.097	.225	.430	.600	-.317	-.406	.171	.459	.683
.150	-.272	-.120	.124	.304	.464	-.313	-.413	-.032	.215	.423
.250	-.259	-.195	.011	.181	.337					
.350	-.289	-.297	-.092	.054	.206					
.450	-.302	-.377	-.201	-.062	.079					
.550	-.264	-.410	-.287	-.165	-.034					
.650	-.199	-.343	-.324	-.224	-.121	-.181	-.183	-.317	-.411	-.297
.750	-.146	-.077	-.236	-.265	-.174	-.270	-.051	-.158	-.264	-.264
.850	-.063	.066	-.025	-.103	-.135	-.239	.062	-.062	-.149	-.202
.900	-.033	.109	.004	-.100	-.165	-.211	.105	-.055	-.158	-.228

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(j) $\alpha = 0°$; M = 0.92

$\dfrac{x}{c_v}$	C_p for – $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for – $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.788	.694	.660	.452	.076	.62?	.813	.681	.368	-.074
.025	.298	-.040	-.448	-.968	-.908	.34?	-.060	-.958	-1.279	-.523
.075	.167	-.084	-.377	-.746	-.726	.17?	-.123	-.669	-1.162	-.531
.150	.061	-.140	-.351	-.636	-.637	.07?	-.144	-.386	-1.063	-.544
.250	.034	-.107	-.224	-.512	-.513	.00?	-.157	-.265	-.569	-.552
.350	-.032	-.140	-.200	-.270	-.452	-.07?	-.231	-.293	-.143	-.541
.450	-.107	-.191	-.208	-.166	-.451	-.16?	-.299	-.286	-.199	-.502
.550	-.117	-.186	-.156	-.126	-.410	-.204	-.324	-.249	-.179	-.451
.650	-.142	-.172	-.105	-.106	-.366	-.237	-.302	-.159	-.160	-.405
.750	-.119	-.076	-.029	-.065	-.312	-.200	-.099	-.076	-.126	-.368
.850	-.029	.028	.045	-.009	-.244	-.050	.037	.014	-.065	-.320
.900	.025	.074	.079	.022	-.206	.024	.093	.067	-.029	-.285
Right side										
.025	-.319	.035	.354	.617	.845	-.895	-.023	.381	.613	.778
.075	-.407	-.085	.162	.373	.586	-.636	-.107	.179	.385	.567
.150	-.369	-.120	.072	.242	.422	-.372	-.123	.087	.254	.418
.250	-.242	-.109	.032	.168	.313	-.254	-.147	.013	.151	.301
.350	-.216	-.147	-.034	.076	.204	-.265	-.198	-.061	.066	.203
.450	-.207	-.191	-.100	-.014	.098	-.246	-.249	-.132	-.037	.093
.550	-.153	-.189	-.117	-.047	.051	-.237	-.296	-.185	-.097	.018
.650	-.092	-.151	-.116	-.074	-.001	-.153	-.271	-.198	-.135	-.034
.750	-.024	-.069	-.095	-.094	-.039	-.063	-.080	-.141	-.141	-.075
.850	.051	.040	.004	-.034	-.030	.026	.053	-.012	-.091	-.089
.900	.082	.074	.034	-.017	-.062	.062	.090	.029	-.064	-.133
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.584	.821	.640	.305	-.081	.801	.894	.701	.359	-.131
.025	.402	-.056	-1.075	-1.122	-.579	.466	-.197	-.950	-.483	-.611
.075	.228	-.107	-.864	-1.018	-.582	.196	-.399	-.403	-.285	-.389
.150	.111	-.147	-.287	-.839	-.594	-.049	-.479	-.449	-.281	-.362
.250	.010	-.198	-.233	-.542	-.591	-.149	-.463	-.391	-.284	-.367
.350	-.099	-.311	-.287	-.372	-.581					
.450	-.217	-.420	-.352	-.285	-.552					
.550	-.308	-.484	-.271	-.217	-.450					
.650	-.387	-.449	-.210	-.239	-.418	-.642	-.216	-.274	-.307	-.403
.750	-.368	-.052	-.150	-.270	-.412	-.378	-.123	-.262	-.315	-.401
.850	-.082	.049	-.095	-.288	-.405	-.116	-.026	-.235	-.305	-.384
.900	-.020	.085	-.065	-.296	-.408	-.079	.019	-.210	-.305	-.374
Right side										
.025	-1.026	-.002	.439	.652	.789	-.939	-.068	.522	.801	.980
.075	-.810	-.090	.228	.429	.604	-.287	-.377	.184	.460	.687
.150	-.231	-.109	.130	.309	.469	-.285	-.417	-.026	.213	.427
.250	-.242	-.193	.018	.185	.343					
.350	-.266	-.295	-.090	.058	.211					
.450	-.303	-.388	-.203	-.061	.080					
.550	-.262	-.437	-.285	-.170	-.033					
.650	-.200	-.445	-.347	-.239	-.125	-.283	-.249	-.513	-.527	-.382
.750	-.157	-.082	-.329	-.322	-.218	-.274	-.164	-.248	-.510	-.340
.850	-.087	.066	-.040	-.233	-.164	-.246	-.036	-.067	-.266	-.239
.900	-.051	.110	.005	-.167	-.195	-.215	.027	-.046	-.173	-.261

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(k) $\alpha = 9.4°$; $M = 0.60$

	C_p for –					C_p for –				
$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.412	.795	.434	-.723	-1.227	.143	.689	.130	-.757	-1.197
.025	-.011	-.124	-.240	-.711	-1.957	.353	-.082	-.749	-1.276	-.950
.075	.034	-.122	-.321	-.521	-1.114	.180	-.115	-.465	-1.170	-.932
.150	.000	-.145	-.312	-.426	-.625	.087	-.122	-.339	-.841	-.886
.250	.003	-.093	-.226	-.299	-.411	.037	-.127	-.283	-.351	-.835
.350	-.020	-.104	-.211	-.253	-.344	-.011	-.142	-.271	-.287	-.708
.450	-.038	-.115	-.202	-.231	-.335	-.050	-.158	-.249	-.276	-.565
.550	-.029	-.095	-.161	-.190	-.277	-.041	-.127	-.195	-.231	-.457
.650	-.016	-.062	-.116	-.165	-.215	-.027	-.091	-.141	-.192	-.379
.750	.007	-.023	-.064	-.095	-.162	-.004	-.037	-.073	-.129	-.319
.850	.037	.028	-.001	-.027	-.095	.041	.026	.010	-.047	-.236
.900	.057	.058	.033	.003	-.049	.062	.058	.046	-.002	-.183
	Right side									
.025	-.172	-.039	.015	.513	.794	-.677	-.046	.402	.606	.718
.075	-.316	-.127	.026	.332	.536	-.438	-.104	.195	.393	.545
.150	-.295	-.131	.010	.232	.388	-.334	-.111	.100	.264	.409
.250	-.227	-.086	.001	.171	.294	-.268	-.118	.046	.166	.303
.350	-.213	-.107	-.019	.084	.190	-.247	-.129	.006	.112	.220
.450	-.202	-.122	-.044	.039	.112	-.229	-.140	-.026	.053	.140
.550	-.150	-.089	-.026	.023	.080	-.184	-.115	-.042	.023	.089
.650	-.104	-.062	-.008	.021	.059	-.127	-.084	-.012	.019	.061
.750	-.054	-.021	.015	.028	.047	-.054	-.021	.015	.028	.041
.850	.016	.040	.055	.057	.059	.030	.046	.064	.059	.043
.900	.043	.062	.064	.066	.054	.053	.060	.064	.057	.011
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.003	.660	-.042	-.453	-.874	.407	.810	-.213	-.308	-.824
.025	.418	-.084	-1.057	-.766	-.609	.587	-.174	-1.199	-.464	-.441
.075	.246	-.104	-.551	-.736	-.612	.339	-.181	-1.017	-.478	-.413
.150	.148	-.111	-.395	-.666	-.595	.139	-.203	-.859	-.471	-.423
.250	.082	-.122	-.328	-.557	-.565	.053	-.167	-.515	-.464	-.432
.350	.021	-.147	-.303	-.467	-.542					
.450	-.022	-.163	-.280	-.405	-.515					
.550	-.054	-.151	-.238	-.369	-.496					
.650	-.036	-.107	-.168	-.335	-.469	-.075	-.131	-.161	-.378	-.402
.750	-.004	-.046	-.089	-.285	-.446	-.025	-.053	-.069	-.331	-.379
.850	.021	.010	-.017	-.219	-.411	.018	.019	-.001	-.276	-.349
.900	.041	.037	.024	-.190	-.397	.046	.058	.042	-.249	-.330
	Right side									
.025	-.945	-.019	.443	.620	.704	-1.097	-.071	.679	.855	.997
.075	-.500	-.075	.251	.440	.580	-.968	-.124	.350	.545	.739
.150	-.350	-.082	.166	.320	.462	-.838	-.115	.195	.357	.527
.250	-.311	-.107	.096	.230	.368					
.350	-.297	-.129	.033	.143	.259					
.450	-.263	-.145	-.012	.078	.170					
.550	-.218	-.140	-.033	.023	.091					
.650	-.150	-.100	-.024	-.002	.031	-.116	-.086	-.021	-.029	-.033
.750	-.072	-.041	.003	-.002	-.015	-.059	-.039	-.010	-.065	-.114
.850	.018	.040	.060	.021	-.052	.028	.042	.048	-.056	-.146
.900	.055	.064	.069	.007	-.086	.055	.062	.060	-.083	-.181

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0^\circ$ - Continued

(l) $\alpha = 9.6^\circ$; $M = 0.80$

$\dfrac{x}{c_v}$	C_p for $z/b_v = 0.11$					C_p for $z/b_v = 0.38$				
	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
	Left side									
.000	.379	.864	.433	-.404	-.214	.417	.740	.433	-.133	-.707
.025	-.026	-.102	-.153	-.566	-1.347	.360	-.076	-.983	-1.143	-.779
.075	.038	-.117	-.313	-.519	-1.009	.187	-.120	-.488	-1.078	-.793
.150	.009	-.154	-.351	-.457	-.719	.102	-.120	-.354	-.898	-.759
.250	.006	-.099	-.244	-.315	-.467	.042	-.130	-.294	-.514	-.719
.350	-.020	-.106	-.224	-.263	-.382	-.022	-.162	-.293	-.316	-.651
.450	-.047	-.132	-.221	-.238	-.343	-.061	-.183	-.279	-.260	-.561
.550	-.041	-.108	-.179	-.196	-.296	-.061	-.153	-.223	-.229	-.472
.650	-.032	-.078	-.133	-.171	-.244	-.049	-.117	-.160	-.206	-.412
.750	-.006	-.034	-.070	-.125	-.189	-.015	-.046	-.075	-.163	-.355
.850	.039	.032	.000	-.045	-.118	.042	.035	.017	-.084	-.287
.900	.067	.068	.038	-.010	-.069	.076	.080	.062	-.032	-.230
	Right side									
.025	-.092	-.007	.001	.425	.800	-1.070	-.028	.411	.622	.747
.075	-.325	-.117	.047	.334	.548	-.444	-.097	.207	.409	.571
.150	-.337	-.126	.030	.246	.399	-.345	-.106	.117	.285	.436
.250	-.249	-.084	.010	.189	.306	-.288	-.120	.056	.193	.327
.350	-.237	-.106	-.020	.109	.201	-.276	-.139	.010	.126	.243
.450	-.235	-.133	-.044	.033	.112	-.273	-.159	-.037	.051	.154
.550	-.180	-.102	-.034	.020	.088	-.226	-.148	-.051	.011	.092
.650	-.124	-.064	-.019	.002	.052	-.151	-.111	-.034	-.004	.055
.750	-.067	-.025	.007	.008	.038	-.058	-.031	.001	.004	.032
.850	.013	.047	.056	.045	.048	.033	.052	.062	.033	.015
.900	.047	.070	.068	.050	.034	.070	.068	.073	.023	-.018

$\dfrac{x}{c_v}$	C_p for $z/b_v = 0.66$					C_p for $z/b_v = 0.93$				
	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
	Left side									
.000	.305	.706	.300	-.075	-.654	.661	.876	.367	.063	-.587
.025	.427	-.085	-1.090	-.633	-.555	.617	-.228	-.941	-.353	-.392
.075	.259	-.097	-.651	-.667	-.571	.326	-.246	-.744	-.375	-.393
.150	.163	-.114	-.394	-.641	-.558	.116	-.264	-.660	-.416	-.395
.250	.093	-.132	-.330	-.525	-.551	.029	-.210	-.605	-.407	-.402
.350	.021	-.177	-.336	-.378	-.518					
.450	-.047	-.214	-.328	-.355	-.488					
.550	-.089	-.207	-.276	-.364	-.464					
.650	-.070	-.150	-.189	-.364	-.447	-.116	-.148	-.206	-.370	-.395
.750	-.025	-.060	-.098	-.343	-.435	-.052	-.037	-.147	-.352	-.373
.850	.021	.014	-.014	-.294		.007	.050	-.075	-.312	-.343
.900	.052	.053	.026	-.266	-.395	.038	.089	-.044	-.290	-.324
	Right side									
.025	-1.035	.005	.472	.650	.759	-.900	-.085	.669	.886	1.063
.075	-.543	-.066	.276	.469	.630	-.804	-.168	.350	.570	.791
.150	-.352	-.069	.186	.354	.518	-.801	-.141	.181	.376	.573
.250	-.349	-.114	.110	.256	.413					
.350	-.366	-.157	.035	.158	.301					
.450	-.343	-.187	-.032	.069	.194					
.550	-.273	-.187	-.072	-.001	.100					
.650	-.180	-.141	-.060	-.039	.031	-.139	-.091	-.061	-.078	-.035
.750	-.083	-.052	-.020	-.052	-.032	-.072	-.033	-.041	-.120	-.143
.850	.024	.047	.050	-.030	-.065	-.003	.062	.020	-.114	-.197
.900	.070	.085	.071	-.048	-.118	.036	.098	.027	-.128	-.247

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = 0° - Continued

(m) α = 9.7°; M = 0.85

	C_p for –					C_p for –				
$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v=0.11$					$z/b_v=0.38$				
	Left side									
.000	.368	.881	.433	-.292		.487	.751	.508	.003	
.025	-.046	-.099	-.094	-.515		.350	-.091	-1.149	-1.220	
.075	.035	-.122	-.297	-.524		.183	-.130	-.442	-1.046	
.150	.006	-.167	-.362	-.505		.096	-.130	-.345	-.834	
.250	.006	-.112	-.242	-.328		.038	-.140	-.287	-.549	
.350	-.029	-.122	-.220	-.272		-.031	-.179	-.280	-.345	
.450	-.059	-.147	-.223	-.245		-.080	-.213	-.274	-.268	
.550	-.052	-.127	-.173	-.200		-.080	-.184	-.222	-.233	
.650	-.048	-.101	-.134	-.176		-.076	-.143	-.162	-.216	
.750	-.019	-.049	-.075	-.140		-.038	-.065	-.085	-.180	
.850	.036	.029	.001	-.066		.032	.027	.010	-.109	
.900	.062	.067	.044	-.022		.066	.077	.062	-.063	
	Right side									
.025	-.041	-.004	-.019	.362		-1.023	-.021	.403	.624	
.075	-.313	-.119	.048	.318		-.422	-.092	.204	.411	
.150	-.341	-.123	.034	.246		-.330	-.106	.118	.285	
.250	-.244	-.089	.011	.189		-.273	-.122	.058	.192	
.350	-.234	-.116	-.021	.112		-.268	-.147	.010	.123	
.450	-.236	-.147	-.053	.033		-.268	-.173	-.048	.046	
.550	-.182	-.113	-.044	.016		-.229	-.170	-.067	.003	
.650	-.123	-.084	-.031	-.010		-.155	-.126	-.057	-.020	
.750	-.073	-.042	-.007	-.007		-.063	-.040	-.019	-.019	
.850	.015	.038	.052	.031		.029	.051	.050	.011	
.900	.044	.070	.065	.037		.066	.073	.062	-.007	
	$z/b_v=0.66$					$z/b_v=0.93$				
	Left side									
.000	.377	.729	.383	-.019		.734	.899	.489	.140	
.025	.417	-.086	-1.144	-.647		.593	-.286	-.831	-.339	
.075	.257	-.103	-.619	-.678		.308	-.282	-.591	-.356	
.150	.160	-.125	-.372	-.658		.095	-.299	-.490	-.405	
.250	.085	-.147	-.300	-.532		.015	-.230	-.469	-.395	
.350	.009	-.208	-.306	-.362						
.450	-.072	-.247	-.311	-.348						
.550	-.122	-.244	-.273	-.371						
.650	-.107	-.170	-.200	-.379		-.170	-.154	-.266	-.371	
.750	-.058	-.066	-.122	-.369		-.096	-.035	-.227	-.356	
.850	-.001	.020	-.047	-.333		-.038	.060	-.172	-.322	
.900	.025	.061	-.006	-.303		-.012	.097	-.139	-.303	
	Right side									
.025	-1.196	.021	.473	.657		-.666	-.085	.659	.894	
.075	-.454	-.055	.277	.469		-.586	-.190	.337	.572	
.150	-.314	-.062	.190	.359		-.560	-.150	.173	.370	
.250	-.301	-.112	.108	.266						
.350	-.331	-.173	.024	.153						
.450	-.331	-.211	-.051	.057						
.550	-.276	-.221	-.104	-.023						
.650	-.192	-.160	-.102	-.062		-.240	-.102	-.101	-.116	
.750	-.107	-.061	-.057	-.089		-.196	-.034	-.084	-.159	
.850	-.004	.050	.027	-.063		-.135	.070	-.024	-.142	
.900	.044	.087	.052	-.084		-.086	.105	-.019	-.170	

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(n) $\alpha = 9.7°$; $M = 0.90$

$\dfrac{x}{c_v}$	C_p for–					C_p for–				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.323	.927	.424	-.203		.549	.761	.572	.144	
.025	-.069	-.077	-.029	-.443		.353	-.081	-1.046	-1.302	
.075	.039	-.106	-.256	-.482		.183	-.129	-.561	-1.023	
.150	.015	-.165	-.405	-.558		.098	-.137	-.385	-.814	
.250	.005	-.118	-.260	-.365		.031	-.153	-.269	-.535	
.350	-.033	-.133	-.226	-.294		-.039	-.203	-.280	-.366	
.450	-.070	-.186	-.242	-.262		-.102	-.258	-.293	-.276	
.550	-.073	-.168	-.195	-.208		-.122	-.256	-.246	-.235	
.650	-.082	-.156	-.151	-.184		-.131	-.221	-.180	-.227	
.750	-.057	-.097	-.091	-.146		-.094	-.116	-.102	-.200	
.850	-.001	-.001	-.013	-.090		-.015	.006	-.005	-.141	
.900	.034	.040	.031	-.046		.031	.059	.046	-.097	
Right side										
.025	.054	.022	-.029	.356		-.935	-.017	.407	.624	
.075	-.265	-.113	.054	.327		-.531	-.097	.211	.413	
.150	-.383	-.134	.043	.258		-.357	-.112	.120	.292	
.250	-.261	-.098	.016	.199		-.266	-.132	.054	.201	
.350	-.243	-.134	-.020	.118		-.273	-.171	.003	.131	
.450	-.259	-.179	-.056	.029		-.290	-.217	-.059	.044	
.550	-.206	-.161	-.056	.010		-.265	-.226	-.097	-.008	
.650	-.150	-.136	-.056	-.022		-.182	-.195	-.099	-.039	
.750	-.094	-.087	-.035	-.031		-.087	-.104	-.059	-.051	
.850	-.007	.011	.026	.009		.007	.030	.014	-.023	
.900	.031	.046	.046	.006		.045	.051	.032	-.042	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.447	.735	.454	.057		.794	.923	.586	.221	
.025	.415	-.085	-1.174	-.719		.586	-.274	-.755	-.334	
.075	.258	-.099	-.969	-.739		.301	-.344	-.501	-.338	
.150	.163	-.125	-.235	-.675		.079	-.343	-.408	-.394	
.250	.082	-.167	-.265	-.537		-.001	-.239	-.389	-.378	
.350	-.013	-.249	-.293	-.325						
.450	-.114	-.311	-.313	-.338						
.550	-.197	-.383	-.283	-.362						
.650	-.210	-.285	-.219	-.376		-.387	-.159	-.289	-.354	
.750	-.155	-.094	-.157	-.378		-.157	-.040	-.273	-.345	
.850	-.066	.004	-.094	-.357		-.103	.047	-.235	-.323	
.900	-.037	.051	-.060	-.337		-.085	.081	-.213	-.313	
Right side										
.025	-1.103	.020	.475	.667		-.541	-.101	.656	.904	
.075	-.802	-.058	.283	.484		-.439	-.264	.339	.588	
.150	-.238	-.070	.197	.370		-.413	-.172	.163	.376	
.250	-.259	-.132	.109	.278						
.350	-.311	-.202	.014	.163						
.450	-.337	-.276	-.079	.052						
.550	-.287	-.315	-.156	-.042						
.650	-.211	-.295	-.178	-.098		-.274	-.156	-.198	-.152	
.750	-.153	-.105	-.133	-.133		-.257	-.051	-.148	-.205	
.850	-.059	.040	-.027	-.104		-.210	.053	-.083	-.197	
.900	-.019	.077	-.004	-.136		-.177	.090	-.083	-.231	

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(o) $\alpha = 9.7°$; $M = 0.92$

$z/b_v = 0.11$ and $z/b_v = 0.38$

x/c_v	\multicolumn{5}{c}{C_p for $z/b_v = 0.11$}	\multicolumn{5}{c}{C_p for $z/b_v = 0.38$}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side
.000	-.337	.980	.292	.448		.179	.749	.559	.590	
.025	-.358	-.050	-.038	.039		-1.279	-.052	.368	-.963	
.075	-.392	-.075	.056	-.162		-1.027	-.099	.199	-.403	
.150	-.484	-.140	.052	-.316		-.725	-.105	.114	-.324	
.250	-.384	-.095	.033	-.220		-.472	-.126	.049	-.262	
.350	-.288	-.117	-.003	-.212		-.341	-.183	-.019	-.271	
.450	-.263	-.171	-.048	-.235		-.279	-.239	-.093	-.298	
.550	-.218	-.165	-.068	-.196		-.252	-.245	-.119	-.254	
.650	-.201	-.159	-.085	-.167		-.245	-.228	-.137	-.194	
.750	-.171	-.096	-.068	-.108		-.230	-.119	-.111	-.117	
.850	-.109	-.002	-.014	-.028		-.171	.010	-.035	-.021	
.900	-.070	.042	.020	.014		-.121	.063	.019	.036	
Right side
.025	.287	.061	.093	.026		.627	.026	-.859	.422	
.075	.354	-.069	-.183	.086		.419	-.052	-.420	.232	
.150	.305	-.105	-.306	.080		.298	-.074	-.320	.137	
.250	.215	-.071	-.227	.059		.203	-.100	-.245	.068	
.350	.115	-.108	-.233	.014		.120	-.145	-.261	.009	
.450	.027	-.159	-.256	-.037		.030	-.192	-.283	-.059	
.550	.000	-.147	-.211	-.058		-.030	-.214	-.270	-.102	
.650	-.043	-.130	-.164	-.069		-.067	-.200	-.195	-.119	
.750	-.058	-.090	-.102	-.049		-.080	-.132	-.098	-.086	
.850	-.020	.013	-.020	.009		-.060	.031	-.001	-.009	
.900	-.020	.043	.018	.030		-.075	.049	.035	.008	

$z/b_v = 0.66$ and $z/b_v = 0.93$

x/c_v	\multicolumn{5}{c}{C_p for $z/b_v = 0.66$}	\multicolumn{5}{c}{C_p for $z/b_v = 0.93$}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side
.000	.078	.724	.439	.451		.248	.921	.789	.592	
.025	-.745	-.054	.429	-1.090		-.346	-.254	.592	-.727	
.075	-.767	-.070	.276	-.881		-.356	-.317	.315	-.490	
.150	-.693	-.100	.178	-.250		-.412	-.300	.091	-.376	
.250	-.495	-.141	.094	-.254		-.400	-.204	.011	-.362	
.350	-.330	-.222	.003	-.286						
.450	-.360	-.292	-.103	-.319						
.550	-.376	-.364	-.191	-.294						
.650	-.389	-.330	-.216	-.227		-.370	-.159	-.474	-.286	
.750	-.392	-.099	-.185	-.165		-.366	-.039	-.203	-.271	
.850	-.376	.009	-.082	-.104		-.345	.048	-.124	-.237	
.900	-.360	.052	-.047	-.070		-.329	.084	-.111	-.215	
Right side
.025	.659	.057	-1.052	.482		.884	-.036	-.507	.659	
.075	.479	-.018	-.617	.298		.570	-.213	-.411	.341	
.150	.373	-.037	-.239	.209		.359	-.128	-.388	.176	
.250	.261	-.100	-.245	.115						
.350	.148	-.168	-.296	.020						
.450	.036	-.250	-.335	-.078						
.550	-.064	-.293	-.295	-.165						
.650	-.134	-.308	-.216	-.200		-.259	-.212	-.275	-.349	
.750	-.191	-.147	-.160	-.194		-.237	-.052	-.265	-.185	
.850	-.138	.051	-.073	-.053		-.212	.056	-.231	-.098	
.900	-.164	.084	-.035	-.021		-.252	.095	-.199	-.102	

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TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = 0° - Continued

(p) α = 15.6°; M = 0.60

$z/b_v = 0.11$ and $z/b_v = 0.38$ — Left side

$\dfrac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	-.351	.823	.179	.080	-.500	.273	.746	.333	-.404	-.850
.025	-.612	-.136	.288	.089	-.480	.137	-.093	-.314	-.668	-.893
.075	-.170	-.143	.019	-.084	-.414	.069	-.132	-.291	-.484	-.767
.150	-.038	-.159	-.126	-.184	-.387	.055	-.129	-.259	-.379	-.627
.250	.012	-.116	-.137	-.170	-.326	.041	-.134	-.246	-.325	-.520
.350	.037	-.100	-.162	-.184	-.288	.023	-.161	-.241	-.291	-.437
.450	.037	-.120	-.178	-.184	-.250	.007	-.172	-.226	-.259	-.372
.550	.037	-.102	-.151	-.147	-.200	.016	-.143	-.187	-.218	-.313
.650	.046	-.077	-.108	-.118	-.153	.016	-.111	-.146	-.179	-.265
.750	.048	-.043	-.069	-.077	-.116	.028	-.061	-.087	-.122	-.202
.850	.048	.003	-.022	-.025	-.065	.046	.000	-.024	-.063	-.139
.900	.062	.037	.005	.003	-.028	-.154	-.184	-.144	-.154	-.157

Right side

$\dfrac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	.350	-.100	-.716	-.254	-.098	-.403	-.088	.109	.193	.204
.075	-.004	-.175	-.198	-.068	.033	-.324	-.136	.046	.121	.240
.150	-.145	-.159	-.056	.012	.109	-.285	-.134	.035	.107	.261
.250	-.158	-.118	.014	.062	.163	-.256	-.134	.041	.114	.281
.350	-.181	-.107	.032	.080	.191	-.240	-.145	.030	.105	.272
.450	-.188	-.125	.037	.087	.182	-.228	-.154	.019	.096	.249
.550	-.156	-.102	.037	.089	.159	-.192	-.141	.010	.084	.211
.650	-.111	-.072	.046	.084	.132	-.136	-.107	.021	.080	.173
.750	-.070	-.036	.046	.078	.105	-.083	-.054	.037	.078	.141
.850	-.004	.023	.062	.080	.091	-.013	.016	.057	.080	.109
.900	.014	.050	.064	.080	.071	.014	.025	.048	.064	.080

$z/b_v = 0.66$ and $z/b_v = 0.93$ — Left side

$\dfrac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	-.009	.675	.060	-.545	-.728	.286	.818	-.178	-.682	-.611
.025	.463	-.068	-.988	-1.129	-.730	.713	-.079	-.954	-.897	-.579
.075	.298	-.091	-.533	-1.020	-.697	.479	-.111	-.836	-.563	-.457
.150	.205	-.104	-.388	-.800	-.656	.254	-.157	-.841	-.545	-.432
.250	.139	-.116	-.318	-.518	-.611	.132	-.154	-.680	-.566	-.441
.350	.089	-.141	-.305	-.382	-.550					
.450	.037	-.161	-.287	-.320	-.489					
.550	.001	-.157	-.241	-.291	-.432					
.650	-.004	-.125	-.171	-.225	-.376	-.054	-.150	-.153	-.329	-.308
.750	.019	-.070	-.097	-.170	-.317	-.029	-.075	-.078	-.254	-.272
.850	.032	-.011	-.026	-.109	-.261	.010	.003	-.013	-.193	-.232
.900	.041	.018	.005	-.075	-.236	.025	.037	.019	-.163	-.218

Right side

$\dfrac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	-1.033	-.038	.471	.621	.615	-.997	-.018	.720	.891	.974
.075	-.582	-.088	.279	.439	.518	-.897	-.107	.446	.630	.784
.150	-.374	-.086	.200	.325	.448	-.859	-.129	.241	.409	.570
.250	-.328	-.109	.139	.253	.385					
.350	-.312	-.136	.080	.175	.326					
.450	-.281	-.147	.041	.130	.263					
.550	-.233	-.141	.014	.089	.213					
.650	-.170	-.111	.010	.066	.157	-.126	-.095	-.011	.032	.060
.750	-.097	-.061	.023	.057	.105	-.074	-.063	-.008	-.020	-.042
.850	.001	.018	.069	.068	.069	.005	.025	.041	-.002	-.080
.900	.032	.041	.069	.057	.028	.030	.057	.057	-.016	-.114

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(q) $\alpha = 15.8°$; M = 0.80

$\frac{x}{c_v}$	C_p for – β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	C_p for – β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.034	.853	.404	.098	-.508	.298	.789	.296	-.226	-.541
.025	-.964	.130	.319	.039	-.463	-.010	-.037	-.304	-.523	-.632
.075	-.568	-.075	-.001	-.125	-.412	-.075	-.126	-.310	-.430	-.576
.150	-.261	-.176	-.190	-.228	-.385	-.056	-.141	-.287	-.370	-.513
.250	-.128	-.129	-.173	-.193	-.327	-.043	-.150	-.276	-.331	-.460
.350	-.061	-.111	-.192	-.193	-.273	-.028	-.177	-.263	-.299	-.404
.450	-.019	-.132	-.207	-.198	-.231	-.034	-.199	-.252	-.267	-.351
.550	.006	-.103	-.158	-.148	-.184	-.011	-.161	-.213	-.231	-.302
.650	.024	-.067	-.116	-.114	-.145	.000	-.128	-.164	-.195	-.258
.750	.034	-.032	-.083	-.083	-.110	.016	-.067	-.104	-.145	-.207
.850	.046	.007	-.034	-.039	-.066	.042	.009	-.036	-.084	-.151
.900	.052	.043	-.002	-.008	-.033	-.143	-.167	-.133	-.136	-.137
	Right side									
.025	.384	-.801	-.939	-.417	-.309	-.401	-.268	-.054	.007	.069
.075	-.025	-.400	-.576	-.240	-.170	-.351	-.253	-.092	-.030	.115
.150	-.202	-.236	-.280	-.151	-.045	-.317	-.188	-.073	-.013	.172
.250	-.199	-.129	-.142	-.073	.072	-.277	-.140	-.036	.022	.231
.350	-.214	-.075	-.063	-.011	.142	-.265	-.121	-.019	.039	.253
.450	-.214	-.061	-.020	.031	.177	-.250	-.126	-.017	.052	.250
.550	-.167	-.032	.007	.051	.169	-.218	-.109	-.007	.063	.231
.650	-.117	-.014	.024	.066	.148	-.161	-.070	.010	.072	.204
.750	-.076	.009	.036	.072	.124	-.096	-.020	.031	.077	.164
.850	-.016	.048	.057	.081	.097	-.022	.043	.058	.084	.130
.900	.001	.060	.060	.075	.080	.003	.052	.051	.067	.086
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.366	.720	.461	.066	-.397	.600	.877	.399	-.140	-.656
.025	.394	-.035	-1.007	-.946	-.645	.656	-.090	-.714	-.848	-.582
.075	.236	-.078	-.504	-.822	-.621	.414	-.167	-.533	-.420	-.434
.150	.155	-.103	-.358	-.657	-.591	.192	-.203	-.483	-.376	-.409
.250	.104	-.120	-.289	-.461	-.553	.072	-.191	-.501	-.389	-.412
.350	.048	-.161	-.289	-.343	-.513					
.450	-.005	-.197	-.301	-.286	-.468					
.550	-.041	-.196	-.273	-.269	-.413					
.650	-.034	-.152	-.201	-.249	-.363	-.117	-.161	-.261	-.358	-.276
.750	-.010	-.073	-.120	-.208	-.312	-.073	-.058	-.208	-.326	-.249
.850	.018	-.003	-.046	-.161	-.271	-.037	.037	-.143	-.276	-.219
.900	.027	.037	-.007	-.131	-.252	-.023	.075	-.107	-.255	-.208
	Right side									
.025	-1.093	-.059	.413	.614	.710	-.730	.016	.669	.889	1.042
.075	-.522	-.093	.225	.401	.579	-.533	-.128	.380	.614	.837
.150	-.338	-.064	.163	.308	.507	-.519	-.132	.186	.383	.614
.250	-.298	-.084	.108	.236	.444					
.350	-.309	-.117	.045	.163	.374					
.450	-.309	-.137	.004	.110	.305					
.550	-.271	-.134	-.025	.074	.247					
.650	-.203	-.097	-.025	.054	.184	-.233	-.093	-.051	-.016	.069
.750	-.120	-.035	-.001	.039	.122	-.197	-.035	-.057	-.072	-.039
.850	-.025	.051	.049	.049	.077	-.128	.063	-.001	-.057	-.098
.900	.016	.077	.063	.039	.027	-.096	.093	.007	-.081	-.157

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Concluded

(r) $\alpha = 15.9°$; M = 0.85

$z/b_v = 0.11$ and $z/b_v = 0.38$

$\frac{x}{c_v}$	C_p for – $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for – $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.410	.905	.585	.091	-.584	.403	.783	.322	-.237	-.517
.025	-1.053	.008	.238	-.095	-.547	.207	-.150	-.425	-.597	-.641
.075	-.489	-.200	-.054	-.225	-.496	.035	-.224	-.409	-.501	-.587
.150	-.227	-.287	-.279	-.323	-.452	-.024	-.209	-.364	-.432	-.531
.250	-.155	-.202	-.231	-.262	-.382	-.061	-.198	-.318	-.377	-.482
.350	-.125	-.152	-.230	-.241	-.315	-.084	-.214	-.295	-.336	-.428
.450	-.102	-.157	-.235	-.231	-.256	-.102	-.231	-.279	-.303	-.375
.550	-.067	-.119	-.176	-.170	-.200	-.081	-.194	-.240	-.258	-.327
.650	-.043	-.084	-.131	-.129	-.156	-.067	-.152	-.192	-.221	-.281
.750	-.013	-.044	-.097	-.105	-.122	-.038	-.089	-.129	-.170	-.225
.850	.018	.005	-.043	-.058	-.075	.010	-.007	-.054	-.110	-.176
.900	.045	.038	-.012	-.028	-.040	-.156	-.166	-.139	-.145	-.141
Right side										
.025	.271	-.704	-1.064	-.478	-.412	-.940	-.270	-.074	-.044	.001
.075	-.159	-.851	-.656	-.313	-.270	-.700	-.306	-.153	-.096	.034
.150	-.436	-.317	-.340	-.238	-.134	-.475	-.253	-.144	-.086	.098
.250	-.298	-.225	-.220	-.163	-.011	-.288	-.200	-.111	-.045	.163
.350	-.246	-.149	-.151	-.089	.072	-.270	-.176	-.088	-.023	.186
.450	-.223	-.120	-.091	-.034	.126	-.264	-.178	-.080	-.006	.203
.550	-.165	-.070	-.043	-.001	.135	-.227	-.152	-.057	.013	.193
.650	-.122	-.040	-.012	.030	.129	-.172	-.102	-.025	.036	.175
.750	-.071	-.009	.011	.043	.112	-.100	-.041	.002	.050	.148
.850	-.007	.036	.042	.058	.091	-.016	.031	.039	.061	.116
.900	.014	.050	.046	.054	.072	.011	.048	.035	.043	.072

$z/b_v = 0.66$ and $z/b_v = 0.93$

$\frac{x}{c_v}$	C_p for – $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for – $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.337	.721	.506	.172	-.261	.635	.892	.507	-.001	-.615
.025	.408	-.088	-1.169	-.966	-.651	.650	-.109	-.748	-.906	-.615
.075	.241	-.118	-.521	-.830	-.624	.405	-.217	-.484	-.410	-.449
.150	.150	-.133	-.360	-.666	-.601	.165	-.248	-.418	-.362	-.412
.250	.085	-.150	-.292	-.489	-.563	.04	-.226	-.428	-.367	-.423
.350	.014	-.195	-.291	-.360	-.521					
.450	-.063	-.242	-.312	-.298	-.475					
.550	-.109	-.246	-.286	-.272	-.426	-.195	-.177	-.301	-.367	-.304
.650	-.105	-.180	-.221	-.262	-.382	-.143	-.064	-.268	-.349	-.276
.750	-.073	-.091	-.151	-.244	-.338	-.097	.039	-.214	-.318	-.244
.850	-.033	-.012	-.078	-.207	-.296	-.087	.073	-.180	-.299	-.232
.900	-.017	.028	-.039	-.180	-.279					
Right side										
.025	-1.335	-.077	.378	.596	.727	-.815	-.007	.647	.882	1.054
.075	-.828	-.123	.187	.382	.578	-.453	-.188	.343	.596	.837
.150	-.274	-.102	.129	.284	.494	-.443	-.188	.139	.353	.605
.250	-.271	-.125	.069	.209	.426					
.350	-.298	-.167	-.005	.128	.356					
.450	-.307	-.187	-.049	.068	.288					
.550	-.280	-.184	-.084	.027	.226	-.279	-.120	-.107	-.065	.062
.650	-.222	-.140	-.077	.010	.172	-.259	-.051	-.111	-.122	-.041
.750	-.158	-.067	-.046	-.003	.114	-.205	.056	-.046	-.112	-.105
.850	-.063	.039	.019	.012	.067	-.168	.091	-.043	-.145	-.173
.900	-.019	.073	.035	-.004	.011					

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = -6°$

(a) $\alpha = -9.4°$; M = 0.60

$z/b_v = 0.11$ (Left columns) — $z/b_v = 0.38$ (Right columns)

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
					Left side					
.000	.724	.836	.562	-.168	-1.028	.460	.824	.548	-.092	-.687
.025	.246	-.112	-.546	-1.005	-2.178	.330	-.094	-.594	-1.159	-1.069
.075	.100	-.130	-.363	-.622	-1.082	.119	-.153	-.450	-.929	-1.053
.150	-.014	-.180	-.336	-.475	-.674	.009	-.185	-.363	-.565	-.954
.250	-.052	-.158	-.244	-.322	-.404	-.082	-.222	-.343	-.420	-.817
.350	-.086	-.164	-.228	-.255	-.317	-.150	-.260	-.359	-.374	-.664
.450	-.130	-.190	-.233	-.241	-.281	-.200	-.299	-.359	-.342	-.543
.550	-.141	-.180	-.201	-.193	-.242	-.200	-.263	-.306	-.294	-.461
.650	-.109	-.142	-.149	-.152	-.212	-.173	-.217	-.244	-.251	-.422
.750	-.077	-.098	-.091	-.104	-.190	-.123	-.153	-.167	-.189	-.381
.850	-.025	-.034	-.032	-.044	-.135	-.050	-.066	-.073	-.118	-.317
.900	.005	.007	.007	-.005	-.091	.000	-.023	-.025	-.072	-.270
					Right side					
.025	-.478	-.078	.283	.541	.776	-.711	-.114	.340	.591	.760
.075	-.414	-.078	.078	.275	.479	-.474	-.183	.109	.311	.491
.150	-.348	-.187	-.012	.132	.304	-.389	-.199	.002	.153	.313
.250	-.269	-.174	-.046	.059	.185	-.346	-.231	-.087	.032	.167
.350	-.246	-.185	-.096	-.026	.082	-.353	-.265	-.149	-.065	.057
.450	-.250	-.208	-.137	-.085	-.007	-.351	-.285	-.196	-.129	-.037
.550	-.209	-.180	-.133	-.104	-.043	-.314	-.267	-.194	-.157	-.091
.650	-.150	-.139	-.107	-.088	-.053	-.239	-.212	-.167	-.147	-.103
.750	-.096	-.089	-.071	-.067	-.046	-.159	-.135	-.105	-.101	-.085
.850	-.011	-.009	.000	-.005	-.009	-.055	-.041	-.012	-.035	-.055
.900	.018	.027	.020	.004	.000	-.014	-.023	-.012	-.030	-.075

$z/b_v = 0.66$ (Left columns) — $z/b_v = 0.93$ (Right columns)

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
					Left side					
.000	.422	.838	.530	.022	-.340	.358	.607	.114	.057	-.098
.025	.340	-.171	-.859	-1.106	-.817	.073	-.909	-1.596	-.551	-.505
.075	.105	-.240	-.598	-1.010	-.852	-.278	-.747	-1.603	-.501	-.496
.150	-.061	-.306	-.550	-.746	-.804	-.417	-.699	-1.295	-.484	-.498
.250	-.178	-.368	-.537	-.526	-.649	-.380	-.557	-.747	-.480	-.507
.350	-.264	-.413	-.537	-.443	-.580					
.450	-.317	-.434	-.521	-.411	-.564					
.550	-.310	-.400	-.461	-.395	-.573					
.650	-.255	-.308	-.359	-.381	-.580	-.307	-.356	-.388	-.507	-.518
.750	-.171	-.217	-.251	-.386	-.573	-.187	-.215	-.254	-.496	-.491
.850	-.105	-.126	-.149	-.368	-.523	-.093	-.098	-.128	-.459	-.448
.900	-.073	-.087	-.101	-.342	-.505	-.055	-.048	-.075	-.434	-.425
					Right side					
.025	-.854	-.176	.354	.598	.751	-1.524	-.861	.082	.360	.609
.075	-.642	-.270	.078	.298	.479	-1.556	-.760	-.295	-.062	.185
.150	-.533	-.308	-.062	.114	.288	-1.312	-.685	-.425	-.237	-.028
.250	-.551	-.384	-.187	-.046	.116					
.350	-.558	-.427	-.274	-.163	-.023					
.450	-.519	-.432	-.317	-.241	-.130					
.550	-.455	-.400	-.304	-.264	-.183					
.650	-.358	-.297	-.247	-.246	-.194	-.332	-.263	-.199	-.230	-.196
.750	-.253	-.210	-.158	-.214	-.185	-.241	-.199	-.162	-.253	-.244
.850	-.125	-.094	-.066	-.124	-.158	-.107	-.075	-.064	-.195	-.206
.900	-.059	-.041	-.030	-.134	-.187	-.050	-.023	-.016	-.205	-.219

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(b) α = -9.6°; M = 0.80

$\frac{x}{c_v}$	\multicolumn{5}{c	}{C_p for - , $z/b_v = 0.11$}	\multicolumn{5}{c	}{C_p for - , $z/b_v = 0.38$}
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°

Left side ($z/b_v = 0.11$ and $z/b_v = 0.38$)

x/c_v	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.000	.847	.883	.714	.372	-.100	.664	.866	.729	.381	-.089
.025	.240	-.102	-.554	-.863	-1.374	.324	-.073	-.942	-1.568	-.904
.075	.104	-.130	-.382	-.784	-1.103	.122	-.148	-.452	-1.345	-.899
.150	-.018	-.192	-.362	-.519	-.851	.014	-.184	-.373	-.398	-.816
.250	-.055	-.167	-.254	-.298	-.606	-.090	-.225	-.350	-.314	-.778
.350	-.102	-.184	-.244	-.238	-.429	-.189	-.314	-.387	-.323	-.622
.450	-.177	-.231	-.266	-.232	-.345	-.284	-.384	-.415	-.326	-.473
.550	-.205	-.233	-.242	-.206	-.273	-.313	-.355	-.362	-.288	-.371
.650	-.186	-.193	-.190	-.162	-.235	-.283	-.292	-.291	-.258	-.320
.750	-.137	-.126	-.122	-.123	-.204	-.197	-.190	-.201	-.212	-.307
.850	-.061	-.046	-.037	-.062	-.151	-.087	-.081	-.093	-.144	-.300
.900	-.016	-.002	.005	-.022	-.118	-.030	-.028	-.033	-.105	-.271

Right side ($z/b_v = 0.11$ and $z/b_v = 0.38$)

x/c_v	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.025	-.424	-.034	.295	.537	.767	-.924	-.081	.349	.580	.758
.075	-.417	-.154	.090	.281	.483	-.456	-.164	.119	.313	.500
.150	-.363	-.187	-.011	.136	.306	-.384	-.199	.005	.157	.321
.250	-.266	-.173	-.053	.054	.184	-.348	-.234	-.090	.028	.167
.350	-.265	-.198	-.116	-.037	.066	-.378	-.296	-.184	-.084	.040
.450	-.278	-.239	-.183	-.126	-.046	-.402	-.357	-.272	-.193	-.097
.550	-.239	-.231	-.195	-.161	-.109	-.367	-.349	-.292	-.250	-.180
.650	-.179	-.181	-.172	-.162	-.130	-.287	-.281	-.265	-.255	-.213
.750	-.116	-.123	-.128	-.135	-.135	-.183	-.167	-.166	-.191	-.191
.850	-.024	-.026	-.029	-.058	-.091	-.069	-.054	-.053	-.103	-.145
.900	.014	.011	-.001	-.032	-.074	-.025	-.028	-.035	-.088	-.166

Left side ($z/b_v = 0.66$ and $z/b_v = 0.93$)

x/c_v	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.000	.677	.901	.749	.417	.025	.658	.739	.603	.452	.184
.025	.348	-.110	-1.093	-1.600	-.899	-.009	-.854	-1.414	-.448	-.426
.075	.121	-.196	-.478	-1.321	-.919	-.333	-.930	-1.315	-.413	-.421
.150	-.043	-.283	-.510	-.633	-.848	-.465	-.848	-1.265	-.412	-.427
.250	-.200	-.395	-.530	-.401	-.582	-.405	-.586	-.851	-.403	-.430
.350	-.345	-.526	-.600	-.327	-.430					
.450	-.465	-.545	-.560	-.344	-.453					
.550	-.519	-.549	-.534	-.370	-.477					
.650	-.427	-.367	-.429	-.388	-.505	-.397	-.408	-.542	-.423	-.462
.750	-.242	-.267	-.318	-.416	-.528	-.268	-.263	-.414	-.430	-.456
.850	-.147	-.155	-.195	-.426	-.521	-.159	-.149	-.279	-.416	-.433
.900	-.107	-.110	-.137	-.429	-.524	-.122	-.107	-.213	-.407	-.427

Right side ($z/b_v = 0.66$ and $z/b_v = 0.93$)

x/c_v	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.025	-1.083	-.102	.371	.612	.777	-1.345	-.605	.021	.360	.636
.075	-.491	-.220	.102	.320	.506	-1.296	-.873	-.379	-.070	.202
.150	-.482	-.278	-.043	.142	.319	-1.232	-.783	-.516	-.277	-.039
.250	-.534	-.401	-.208	-.050	.123					
.350	-.635	-.539	-.345	-.215	-.062					
.450	-.554	-.517	-.444	-.353	-.221					
.550	-.524	-.525	-.470	-.435	-.327					
.650	-.435	-.431	-.428	-.429	-.359	-.432	-.349	-.292	-.336	-.304
.750	-.305	-.263	-.227	-.292	-.301	-.403	-.272	-.231	-.302	-.301
.850	-.158	-.114	-.105	-.190	-.230	-.254	-.131	-.120	-.223	-.230
.900	-.099	-.067	-.065	-.196	-.251	-.185	-.076	-.084	-.226	-.242

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = -6°$ - Continued

(c) $\alpha = -9.7°$; $M = 0.85$

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.881	.906	.760	.478	.053	.707	.888	.772	-.498	.045
.025	.245	-.104	-.539	-.827	-1.312	.328	-.077	-.987	-1.088	-1.080
.075	.110	-.134	-.420	-.784	-1.035	.129	-.161	-.397	-1.217	-1.026
.150	-.016	-.205	-.387	-.662	-.815	.014	-.189	-.373	-1.074	-.726
.250	-.053	-.178	-.259	-.354	-.633	-.085	-.236	-.342	-.337	-.735
.350	-.106	-.193	-.247	-.262	-.442	-.192	-.308	-.388	-.309	-.670
.450	-.187	-.254	-.285	-.249	-.356	-.313	-.421	-.447	-.326	-.528
.550	-.227	-.280	-.266	-.223	-.302	-.366	-.456	-.418	-.346	-.412
.650	-.225	-.249	-.208	-.186	-.262	-.359	-.388	-.309	-.309	-.349
.750	-.171	-.169	-.135	-.132	-.223	-.248	-.225	-.214	-.214	-.308
.850	-.078	-.070	-.044	-.058	-.165	-.111	-.101	-.108	-.113	-.285
.900	-.032	-.020	-.001	-.018	-.129	-.046	-.034	-.042	-.055	-.265
	Right side									
.025	-.412	-.021	.305	.541	.772	-.953	-.074	.356	.579	.760
.075	-.433	-.158	.094	.278	.487	-.470	-.163	.126	.314	.504
.150	-.387	-.196	-.006	.138	.308	-.384	-.192	.013	.162	.325
.250	-.277	-.181	-.054	.055	.192	-.347	-.232	-.088	.037	.179
.350	-.273	-.215	-.119	-.035	.069	-.383	-.308	-.188	-.084	.047
.450	-.293	-.269	-.200	-.137	-.053	-.440	-.394	-.301	-.206	-.100
.550	-.263	-.283	-.235	-.184	-.127	-.416	-.442	-.366	-.288	-.203
.650	-.198	-.233	-.224	-.197	-.168	-.304	-.365	-.359	-.319	-.271
.750	-.129	-.162	-.169	-.176	-.192	-.202	-.192	-.237	-.242	-.279
.850	-.035	-.048	-.055	-.072	-.138	-.081	-.074	-.068	-.102	-.209
.900	.005	-.010	-.021	-.037	-.118	-.045	-.047	-.052	-.068	-.204
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.716	.923	.794	.519	.138	.697	.780	.677	.455	.223
.025	.356	-.095	-1.068	-1.427	-.798	.014	-.725	-1.316	-.994	-.419
.075	.137	-.183	-.781	-1.274	-.815	-.324	-.859	-1.373	-.933	-.408
.150	-.035	-.267	-.468	-1.160	-.800	-.434	-.803	-1.140	-.949	-.409
.250	-.191	-.369	-.502	-.852	-.764	-.459	-.681	-.784	-.966	-.417
.350	-.346	-.514	-.587	-.278	-.548					
.450	-.499	-.638	-.672	-.492	-.436					
.550	-.578	-.668	-.502	-.579	-.439					
.650	-.585	-.627	-.448	-.404	-.462	-.598	-.425	-.574	-.474	-.448
.750	-.300	-.205	-.359	-.369	-.484	-.298	-.340	-.474	-.519	-.448
.850	-.152	-.178	-.227	-.292	-.494	-.199	-.237	-.380	-.484	-.438
.900	-.131	-.138	-.169	-.244	-.504	-.167	-.183	-.310	-.449	-.432
	Right side									
.025	-1.061	-.075	.386	.617	.781	-1.270	-.519	.047	.403	.670
.075	-.740	-.195	.121	.332	.521	-1.250	-.824	-.357	-.042	.226
.150	-.452	-.250	-.021	.153	.333	-1.198	-.719	-.471	-.251	-.016
.250	-.523	-.379	-.187	-.038	.139					
.350	-.628	-.526	-.354	-.207	-.049					
.450	-.654	-.655	-.503	-.370	-.211					
.550	-.490	-.636	-.584	-.468	-.349					
.650	-.475	-.534	-.549	-.525	-.458	-.463	-.455	-.566	-.689	-.736
.750	-.341	-.286	-.401	-.503	-.531	-.467	-.345	-.315	-.322	-.525
.850	-.185	-.139	-.106	-.108	-.279	-.340	-.209	-.166	-.176	-.219
.900	-.125	-.100	-.081	-.101	-.252	-.271	-.155	-.123	-.162	-.213

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(d) α = -9.8°; M = 0.90

$\frac{x}{c_v}$	C_p for — $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for — $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.913	.931	.807	.556	.126	.752	.902	.807	.575	.136
.025	.257	-.081	-.473	-.803	-1.143	.339	-.049	-.887	-1.188	-.691
.075	.122	-.114	-.455	-.732	-.946	.143	-.142	-.689	-1.092	-.691
.150	-.005	-.195	-.415	-.672	-.782	.035	-.170	-.420	-1.004	-.705
.250	-.042	-.163	-.270	-.616	-.602	-.062	-.203	-.290	-.822	-.702
.350	-.092	-.175	-.233	-.325	-.539	-.171	-.281	-.360	-.444	-.653
.450	-.182	-.245	-.273	-.254	-.512	-.300	-.384	-.425	-.215	-.587
.550	-.260	-.308	-.302	-.182	-.433	-.376	-.467	-.473	-.211	-.522
.650	-.291	-.323	-.271	-.167	-.354	-.457	-.523	-.456	-.283	-.446
.750	-.332	-.304	-.171	-.140	-.284	-.486	-.508	-.246	-.259	-.356
.850	-.178	-.096	-.057	-.080	-.221	-.297	-.100	-.104	-.143	-.325
.900	-.078	-.025	-.009	-.039	-.196	-.107	-.013	-.040	-.080	-.304
	Right side									
.025	-.366	-.003	.317	.554	.800	-.869	-.057	.356	.580	.772
.075	-.467	-.140	.101	.294	.515	-.687	-.154	.141	.322	.522
.150	-.438	-.193	-.001	.150	.335	-.483	-.175	.028	.181	.351
.250	-.291	-.167	-.042	.071	.213	-.293	-.209	-.068	.050	.198
.350	-.257	-.195	-.106	-.024	.089	-.359	-.284	-.171	-.065	.070
.450	-.289	-.254	-.193	-.128	-.033	-.435	-.375	-.288	-.199	-.080
.550	-.303	-.312	-.261	-.199	-.120	-.485	-.467	-.376	-.298	-.191
.650	-.254	-.303	-.282	-.244	-.181	-.432	-.517	-.445	-.377	-.273
.750	-.165	-.296	-.351	-.329	-.272	-.228	-.472	-.476	-.436	-.349
.850	-.051	-.068	-.158	-.210	-.257	-.094	-.037	-.331	-.359	-.357
.900	-.008	-.020	-.070	-.132	-.316	-.046	-.012	-.081	-.250	-.430
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.756	.938	.838	.587	.197	.748	.822	.731	.531	.254
.025	.377	-.056	-.965	-1.252	-.782	.075	-.597	-1.126	-.894	-.436
.075	.169	-.146	-.808	-1.124	-.795	-.262	-.753	-1.322	-.838	-.426
.150	.004	-.229	-.666	-1.057	-.819	-.396	-.747	-1.169	-.835	-.426
.250	-.149	-.327	-.386	-1.009	-.924	-.434	-.734	-.839	-.846	-.432
.350	-.307	-.460	-.519	-.746	-.569					
.450	-.462	-.602	-.614	-.307	-.445					
.550	-.601	-.703	-.628	-.449	-.450					
.650	-.659	-.738	-.600	-.485	-.468	-.847	-.717	-.617	-.350	-.450
.750	-.704	-.711	-.427	-.387	-.485	-.774	-.469	-.524	-.504	-.453
.850	-.610	-.092	-.209	-.339	-.490	-.451	-.308	-.461	-.532	-.449
.900	-.260	-.033	-.177	-.306	-.505	-.340	-.261	-.428	-.514	-.446
	Right side									
.025	-.952	-.045	.404	.623	.795	-1.181	-.459	.101	.439	.716
.075	-.806	-.167	.150	.348	.539	-1.226	-.749	-.312	-.005	.282
.150	-.632	-.218	.011	.174	.363	-1.115	-.666	-.392	-.222	.037
.250	-.403	-.334	-.153	-.002	.170					
.350	-.561	-.472	-.314	-.183	-.012					
.450	-.636	-.621	-.459	-.335	-.173					
.550	-.631	-.695	-.586	-.488	-.310					
.650	-.584	-.717	-.674	-.553	-.430	-.515	-.667	-.783	-.797	-.693
.750	-.424	-.663	-.702	-.649	-.550	-.533	-.480	-.707	-.850	-.746
.850	-.187	-.046	-.617	-.648	-.557	-.436	-.281	-.404	-.764	-.738
.900	-.142	-.012	-.197	-.611	-.583	-.392	-.231	-.304	-.579	-.738

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(e) α = -9.8°; M = 0.92

$\dfrac{x}{c_v}$	C_p for $\beta=-3.9°$	C_p for $\beta=0°$	C_p for $\beta=3.9°$	C_p for $\beta=7.9°$	C_p for $\beta=12.7°$	C_p for $\beta=-3.9°$	C_p for $\beta=0°$	C_p for $\beta=3.9°$	C_p for $\beta=7.9°$	C_p for $\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.924	.942	.815	.555	.182	.753	.908	.814	.569	.172
.025	.283	-.082	-.476	-.859	-1.189	.355	-.053	-.885	-1.178	-.912
.075	.143	-.116	-.489	-.746	-.941	.159	-.149	-.737	-1.088	-.891
.150	.011	-.199	-.455	-.700	-.839	.045	-.174	-.509	-1.019	-.777
.250	-.027	-.168	-.436	-.692	-.659	-.049	-.207	-.469	-.932	-.735
.350	-.082	-.182	-.246	-.420	-.519	-.156	-.290	-.336	-.701	-.658
.450	-.173	-.247	-.277	-.315	-.450	-.281	-.380	-.411	-.322	-.560
.550	-.258	-.314	-.318	-.187	-.380	-.367	-.473	-.473	-.149	-.486
.650	-.291	-.342	-.300	-.150	-.342	-.447	-.548	-.463	-.219	-.434
.750	-.355	-.390	-.215	-.142	-.312	-.486	-.566	-.331	-.260	-.392
.850	-.287	-.199	-.104	-.098	-.271	-.480	-.321	-.195	-.188	-.396
.900	-.145	-.062	-.045	-.061	-.244	-.284	-.063	-.086	-.131	-.374
Right side										
.025	-.370	.000	.323	.573	.794	-.865	-.057	.366	.595	.770
.075	-.502	-.149	.109	.315	.514	-.736	-.159	.137	.340	.523
.150	-.460	-.199	-.001	.167	.335	-.557	-.178	.030	.194	.352
.250	-.383	-.172	-.041	.083	.213	-.336	-.207	-.061	.064	.202
.350	-.258	-.196	-.111	-.015	.086	-.329	-.288	-.169	-.051	.073
.450	-.280	-.260	-.196	-.121	-.036	-.410	-.368	-.287	-.192	-.075
.550	-.304	-.329	-.274	-.199	-.126	-.480	-.472	-.390	-.285	-.185
.650	-.263	-.319	-.290	-.240	-.182	-.391	-.533	-.447	-.368	-.273
.750	-.186	-.390	-.385	-.343	-.283	-.297	-.536	-.485	-.424	-.342
.850	-.070	-.161	-.315	-.303	-.266	-.128	-.271	-.520	-.426	-.367
.900	-.026	-.054	-.232	-.277	-.349	-.066	-.023	-.468	-.449	-.435
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.758	.946	.843	.589	.224	.761	.833	.744	.541	.272
.025	.397	-.052	-.959	-1.232	-.868	.110	-.541	-1.129	-.878	-.467
.075	.186	-.141	-.814	-1.109	-.870	-.227	-.730	-1.332	-.833	-.455
.150	.023	-.222	-.760	-1.053	-.839	-.367	-.728	-1.216	-.830	-.454
.250	-.123	-.319	-.634	-1.028	-.714	-.410	-.738	-.850	-.834	-.460
.350	-.281	-.448	-.451	-.957	-.503					
.450	-.437	-.591	-.567	-.449	-.481					
.550	-.583	-.709	-.506	-.394	-.490					
.650	-.650	-.763	-.588	-.463	-.506	-.873	-.783	-.644	-.328	-.482
.750	-.717	-.763	-.626	-.403	-.524	-.918	-.577	-.584	-.509	-.484
.850	-.684	-.541	-.364	-.378	-.527	-.661	-.418	-.544	-.561	-.477
.900	-.662	-.138	-.221	-.377	-.535	-.485	-.346	-.510	-.555	-.476
Right side										
.025	-.948	-.041	.418	.635	.797	-1.119	-.393	.130	.466	.725
.075	-.814	-.167	.158	.362	.541	-1.141	-.709	-.293	.021	.284
.150	-.719	-.213	.024	.195	.364	-1.071	-.667	-.376	-.200	.042
.250	-.587	-.326	-.140	.013	.178					
.350	-.476	-.462	-.303	-.162	-.004					
.450	-.574	-.613	-.449	-.315	-.171					
.550	-.514	-.701	-.588	-.463	-.304					
.650	-.599	-.751	-.683	-.535	-.418	-.516	-.701	-.871	-.787	-.688
.750	-.558	-.730	-.721	-.636	-.541	-.575	-.598	-.910	-.853	-.737
.850	-.238	-.436	-.710	-.647	-.555	-.485	-.380	-.689	-.826	-.728
.900	-.154	-.049	-.681	-.665	-.587	-.451	-.309	-.497	-.808	-.734

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = -6°$ - Continued

(f) $\alpha = 0°$; $M = 0.60$

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.611	.605	.664	-.455	-1.262	.287	.286	.751	-.386	-.987
.025	.277	.275	-.103	-1.117	-1.934	.344	.339	-.091	-1.057	-.987
.075	.130	.132	-.117	-.675	-1.470	.151	.153	-.139	-.927	-.971
.150	.031	.036	-.144	-.474	-.812	.048	.050	-.158	-.682	-.890
.250	.015	.013	-.110	-.300	-.422	-.024	-.026	-.171	-.420	-.823
.350	-.033	-.042	-.128	-.240	-.345	-.088	-.085	-.206	-.344	-.715
.450	-.079	-.079	-.151	-.217	-.306	-.127	-.127	-.222	-.302	-.595
.550	-.081	-.079	-.133	-.170	-.308	-.132	-.129	-.194	-.258	-.486
.650	-.065	-.062	-.094	-.131	-.251	-.113	-.108	-.155	-.221	-.422
.750	-.038	-.037	-.043	-.089	-.172	-.065	-.069	-.094	-.166	-.350
.850	.008	.006	.016	-.029	-.096	.002	-.001	-.009	-.092	-.271
.900	.031	.034	.046	-.001	-.050	.027	.029	.030	-.041	-.214
Right side										
.025	-.481	-.480	-.028	.610	.821	-.661	-.659	-.069	.607	.731
.075	-.403	-.407	-.119	.350	.542	-.465	-.466	-.128	.362	.523
.150	-.327	-.317	-.133	.214	.371	-.369	-.368	-.149	.216	.375
.250	-.238	-.239	-.117	.128	.257	-.318	-.324	-.167	.110	.239
.350	-.221	-.214	-.135	.054	.158	-.316	-.306	-.192	.033	.154
.450	-.212	-.207	-.153	-.013	.080	-.258	-.260	-.180	-.048	.038
.550	-.159	-.161	-.126	-.025	.054	-.263	-.260	-.196	-.071	.017
.650	-.111	-.113	-.091	-.015	.040	-.192	-.186	-.146	-.059	-.001
.750	-.061	-.049	-.041	.001	.036	-.109	-.101	-.078	-.034	-.001
.850	.020	.018	.025	.038	.057	-.012	-.007	.018	.017	.010
.900	.048	.041	.052	.050	.052	.022	.027	.036	.017	-.013
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.185	.188	.742	-.305	-.650	.358	.364	.660	.045	-.382
.025	.374	.373	-.142	-.974	-.763	.266	.272	-.738	-.490	-.447
.075	.162	.165	-.183	-.904	-.805	-.038	-.033	-.528	-.457	-.454
.150	.038	.041	-.226	-.738	-.821	-.212	-.202	-.507	-.444	-.454
.250	-.065	-.060	-.265	-.481	-.740	-.240	-.232	-.457	-.448	-.472
.350	-.148	-.138	-.299	-.388	-.555					
.450	-.192	-.189	-.313	-.365	-.519					
.550	-.194	-.193	-.285	-.360	-.530					
.650	-.159	-.159	-.219	-.358	-.539	-.185	-.177	-.224	-.439	-.491
.750	-.109	-.111	-.119	-.351	-.525	-.107	-.099	-.110	-.432	-.456
.850	-.028	-.028	-.048	-.321	-.468	-.026	-.033	-.021	-.388	-.410
.900	-.008	-.010	-.014	-.295	-.447	.008	.013	.025	-.363	-.389
Right side										
.025	-.900	-.881	-.089	.617	.719	-1.394	-1.395	-.496	.580	.816
.075	-.587	-.583	-.164	.381	.528	-1.375	-1.377	-.505	.172	.408
.150	-.472	-.468	-.187	.228	.382	-1.152	-1.143	-.445	.008	.204
.250	-.470	-.466	-.256	.087	.239					
.350	-.463	-.468	-.288	-.018	.114					
.450	-.421	-.416	-.292	-.094	.022					
.550	-.355	-.351	-.270	-.131	-.043	-.182	-.175	-.185	-.161	-.144
.650	-.261	-.257	-.203	-.129	-.073	-.123	-.122	-.105	-.182	-.172
.750	-.157	-.150	-.123	-.108	-.089	-.021	-.017	.000	-.143	-.156
.850	-.040	-.037	-.018	-.069	-.098	.025	.027	.050	-.157	-.179
.900	.006	.009	.018	-.078	-.128					

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(g) α = 0°; M = 0.80

$\frac{x}{c_v}$	C_p for $z/b_v = 0.11$					C_p for $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side					Left side				
.000	.717	.677	.545	.202	-.275	.498	.780	.561	.088	-.361
.025	.279	-.076	-.521	-1.111	-1.021	.339	-.084	-.900	-1.241	-.663
.075	.146	-.101	-.365	-.828	-.886	.157	-.137	-.431	-1.048	-.670
.150	.042	-.148	-.323	-.463	-.790	.057	-.157	-.333	-.811	-.687
.250	.016	-.116	-.219	-.281	-.637	-.027	-.186	-.291	-.480	-.687
.350	-.047	-.146	-.200	-.228	-.531	-.104	-.241	-.303	-.299	-.624
.450	-.102	-.185	-.202	-.203	-.471	-.174	-.284	-.303	-.240	-.531
.550	-.108	-.165	-.158	-.162	-.382	-.183	-.263	-.252	-.217	-.445
.650	-.101	-.128	-.110	-.127	-.295	-.166	-.208	-.190	-.217	-.367
.750	-.065	-.071	-.051	-.091	-.223	-.108	-.111	-.112	-.188	-.303
.850	-.004	.005	.018	-.038	-.140	-.018	-.007	-.017	-.121	-.254
.900	.028	.046	.054	-.001	-.091	.025	.042	.032	-.071	-.226
	Right side					Right side				
.025	-.416	.002	.331	.599	.818	-.957	-.042	.372	.605	.745
.075	-.393	-.101	.138	.356	.550	-.428	-.122	.161	.371	.535
.150	-.320	-.131	.051	.219	.385	-.347	-.145	.062	.233	.380
.250	-.231	-.117	.012	.143	.275	-.292	-.174	-.019	.122	.261
.350	-.217	-.150	-.051	.058	.169	-.300	-.220	-.091	.032	.160
.450	-.209	-.182	-.107	-.019	.072	-.269	-.231	-.141	-.056	.045
.550	-.166	-.159	-.110	-.044	.034	-.266	-.255	-.181	-.098	-.009
.650	-.105	-.117	-.091	-.048	.008	-.185	-.189	-.153	-.106	-.038
.750	-.042	-.061	-.057	-.039	-.006	-.097	-.091	-.086	-.077	-.047
.850	.028	.020	.015	.011	.011	.004	.013	.000	-.024	-.047
.900	.063	.054	.036	.026	.005	.037	.046	.022	-.019	-.073

$\frac{x}{c_v}$	C_p for $z/b_v = 0.66$					C_p for $z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side					Left side				
.000	.461	.777	.514	.161	-.364	.648	.742	.555	.403	-.107
.025	.370	-.111	-1.040	-.694	-.765	.175	-.750	-.576	-.374	-.381
.075	.178	-.165	-.521	-.702	-.773	-.091	-.598	-.529	-.354	-.385
.150	.042	-.222	-.411	-.656	-.776	-.286	-.592	-.492	-.352	-.387
.250	-.079	-.295	-.368	-.457	-.765	-.300	-.535	-.474	-.355	-.395
.350	-.189	-.385	-.378	-.330	-.509					
.450	-.281	-.436	-.376	-.340	-.436					
.550	-.304	-.402	-.332	-.367	-.437					
.650	-.249	-.258	-.257	-.389	-.459	-.272	-.208	-.335	-.387	-.434
.750	-.148	-.134	-.188	-.395	-.474	-.165	-.081	-.327	-.378	-.428
.850	-.059	-.029	-.123	-.371	-.468	-.099	.017	-.288	-.357	-.405
.900	-.039	.013	-.084	-.348	-.459	-.076	.057	-.245	-.337	-.392
	Right side					Right side				
.025	-1.073	-.047	.403	.619	.742	-.629	-.578	.282	.570	.849
.075	-.506	-.143	.175	.382	.553	-.592	-.641	-.110	.164	.442
.150	-.396	-.182	.059	.243	.408	-.607	-.517	-.225	-.003	.226
.250	-.399	-.280	-.068	.102	.265					
.350	-.425	-.361	-.182	-.032	.131					
.450	-.408	-.411	-.269	-.140	.008					
.550	-.338	-.367	-.294	-.197	-.078					
.650	-.254	-.251	-.242	-.208	-.128	-.246	-.183	-.197	-.217	-.192
.750	-.171	-.128	-.152	-.173	-.145	-.286	-.068	-.165	-.219	-.214
.850	-.067	.005	-.042	-.112	-.130	-.238	.040	-.084	-.158	-.182
.900	-.024	.053	-.013	-.114	-.162	-.200	.080	-.071	-.161	-.194

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = -6°$ - Continued

(h) $\alpha = 0°$; $M = 0.85$

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.752	.687	.585	.301	-.120	.553	.794	.603	.203	-.240
.025	.282	-.058	-.510	-1.037	-.993	.334	-.072	-1.051	-1.512	-.624
.075	.149	-.101	-.375	-.834	-.833	.153	-.132	-.424	-1.320	-.627
.150	.034	-.154	-.345	-.604	-.736	.053	-.152	-.348	-.541	-.651
.250	.004	-.122	-.236	-.271	-.604	-.034	-.188	-.306	-.344	-.656
.350	-.054	-.155	-.209	-.226	-.526	-.121	-.248	-.313	-.306	-.610
.450	-.124	-.195	-.219	-.219	-.491	-.214	-.322	-.315	-.272	-.543
.550	-.147	-.196	-.176	-.178	-.421	-.247	-.338	-.266	-.295	-.474
.650	-.150	-.175	-.127	-.146	-.340	-.244	-.275	-.204	-.221	-.404
.750	-.108	-.094	-.057	-.110	-.268	-.173	-.136	-.127	-.189	-.334
.850	-.037	-.005	.014	-.048	-.183	-.063	-.012	-.035	-.120	-.277
.900	-.001	.046	.044	-.010	-.136	-.007	.042	.014	-.074	-.246
Right side										
.025	-.386	.009	.334	.601	.830	-.943	-.051	.370	.601	.754
.075	-.398	-.105	.139	.355	.563	-.467	-.122	.159	.365	.543
.150	-.344	-.141	.048	.219	.393	-.351	-.146	.063	.229	.392
.250	-.247	-.126	.011	.146	.284	-.301	-.179	-.024	.120	.270
.350	-.224	-.161	-.058	.052	.173	-.313	-.232	-.102	.031	.162
.450	-.227	-.201	-.124	-.034	.070	-.281	-.268	-.170	-.074	.037
.550	-.181	-.192	-.140	-.068	.026	-.273	-.315	-.230	-.132	-.024
.650	-.126	-.151	-.133	-.077	-.011	-.201	-.251	-.214	-.150	-.067
.750	-.054	-.079	-.095	-.068	-.031	-.107	-.112	-.137	-.120	-.078
.850	.022	.016	-.015	-.014	-.013	-.016	.010	-.031	-.056	-.071
.900	.057	.052	.015	.003	-.021	.024	.050	-.001	-.047	-.103
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.516	.792	.565	.180	-.253	.710	.794	.615	.439	-.027
.025	.372	-.087	-1.131	-.797	-.736	.214	-.659	-.467	-.369	-.383
.075	.183	-.145	-.437	-.800	-.747	-.097	-.574	-.454	-.353	-.384
.150	.047	-.202	-.414	-.702	-.751	-.294	-.649	-.420	-.346	-.388
.250	-.081	-.285	-.349	-.502	-.776	-.276	-.619	-.407	-.353	-.397
.350	-.204	-.403	-.362	-.298	-.506					
.450	-.324	-.510	-.368	-.316	-.433					
.550	-.414	-.537	-.326	-.356	-.433					
.650	-.427	-.365	-.267	-.385	-.451	-.401	-.241	-.346	-.387	-.427
.750	-.220	-.126	-.214	-.399	-.467	-.210	-.118	-.344	-.386	-.426
.850	-.103	-.022	-.157	-.390	-.464	-.146	-.014	-.316	-.364	-.410
.900	-.070	.022	-.128	-.379	-.464	-.126	.032	-.298	-.351	-.400
Right side										
.025	-1.094	-.041	.407	.630	.756	-.421	-.523	.288	.592	.867
.075	-.397	-.135	.180	.394	.564	-.411	-.687	-.107	.182	.452
.150	-.394	-.172	.065	.252	.419	-.428	-.594	-.219	.002	.236
.250	-.370	-.282	-.067	.107	.273					
.350	-.381	-.405	-.204	-.035	.132					
.450	-.381	-.487	-.313	-.159	-.004					
.550	-.327	-.526	-.368	-.248	-.107					
.650	-.258	-.332	-.377	-.283	-.168	-.338	-.248	-.332	-.330	-.231
.750	-.198	-.132	-.223	-.241	-.191	-.333	-.116	-.213	-.235	-.243
.850	-.126	.009	-.075	-.137	-.151	-.304	.016	-.113	-.162	-.194
.900	-.090	.060	-.044	-.137	-.180	-.278	.065	-.102	-.166	-.204

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = -6°$ - Continued

(i) $\alpha = 0°$; M = 0.90

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
					Left side					
.000	.780	.693	.638	.407	.018	.603	.808	.664	.327	-.137
.025	.300	-.042	-.458	-.958	-.986	.349	-.060	-.944	-1.309	-.588
.075	.167	-.085	-.367	-.766	-.806	.172	-.129	-.499	-1.170	-.599
.150	.053	-.146	-.354	-.628	-.700	.074	-.149	-.323	-1.038	-.620
.250	.027	-.117	-.218	-.368	-.573	-.010	-.175	-.284	-.217	-.631
.350	-.040	-.149	-.202	-.188	-.502	-.106	-.238	-.307	-.291	-.588
.450	-.115	-.208	-.215	-.194	-.472	-.205	-.323	-.304	-.247	-.529
.550	-.146	-.220	-.174	-.170	-.414	-.251	-.370	-.273	-.223	-.464
.650	-.178	-.220	-.132	-.148	-.351	-.286	-.388	-.203	-.225	-.409
.750	-.158	-.134	-.054	-.117	-.284	-.281	-.180	-.120	-.209	-.343
.850	-.071	-.011	.022	-.054	-.208	-.122	-.002	-.027	-.150	-.291
.900	-.012	.044	.058	-.018	-.161	-.033	.051	.021	-.096	-.254
					Right side					
.025	-.337	.021	.350	.614	.847	-.896	-.038	.383	.602	.769
.075	-.401	-.094	.156	.365	.580	-.561	-.119	.175	.374	.556
.150	-.349	-.134	.060	.233	.412	-.352	-.139	.076	.241	.409
.250	-.235	-.123	.025	.156	.302	-.271	-.169	-.005	.132	.280
.350	-.216	-.158	-.045	.062	.187	-.301	-.224	-.084	.043	.176
.450	-.219	-.211	-.121	-.030	.084	-.266	-.278	-.164	-.068	.051
.550	-.173	-.224	-.151	-.076	.023	-.275	-.351	-.242	-.143	-.024
.650	-.121	-.204	-.160	-.099	-.017	-.194	-.378	-.283	-.185	-.079
.750	-.051	-.121	-.144	-.116	-.058	-.103	-.158	-.223	-.188	-.109
.850	.030	.002	-.037	-.053	-.044	-.014	.008	-.078	-.126	-.117
.900	.065	.044	-.001	-.033	-.063	.025	.049	-.023	-.103	-.147
	$z/b_v = 0.66$					$z/b_v = 0.93$				
					Left side					
.000	.564	.809	.622	.280	-.148	.749	.836	.681	.493	.057
.025	.390	-.064	-1.045	-.805	-.693	.246	-.543	-.652	-.348	-.373
.075	.205	-.122	-.821	-.786	-.697	-.052	-.536	-.355	-.334	-.374
.150	.073	-.176	-.323	-.690	-.702	-.263	-.609	-.343	-.330	-.374
.250	-.047	-.250	-.342	-.452	-.758	-.252	-.635	-.346	-.334	-.383
.350	-.178	-.380	-.308	-.311	-.538					
.450	-.313	-.507	-.367	-.322	-.420					
.550	-.389	-.592	-.316	-.352	-.412					
.650	-.483	-.588	-.249	-.379	-.428	-.774	-.280	-.320	-.367	-.408
.750	-.508	-.104	-.205	-.394	-.441	-.591	-.207	-.321	-.369	-.409
.850	-.212	.000	-.160	-.391	-.447	-.199	-.134	-.303	-.352	-.404
.900	-.088	.031	-.143	-.387	-.451	-.161	-.102	-.285	-.344	-.398
					Right side					
.025	-1.022	-.021	.430	.639	.771	-.299	-.447	.323	.621	.882
.075	-.768	-.118	.207	.407	.586	-.326	-.647	-.074	.207	.481
.150	-.299	-.150	.096	.268	.440	-.333	-.577	-.185	.029	.261
.250	-.385	-.247	-.040	.125	.294					
.350	-.310	-.384	-.175	-.018	.151					
.450	-.362	-.497	-.288	-.152	.010					
.550	-.307	-.559	-.417	-.250	-.106					
.650	-.247	-.592	-.445	-.337	-.188	-.315	-.343	-.707	-.643	-.471
.750	-.200	-.142	-.477	-.379	-.283	-.313	-.269	-.454	-.627	-.482
.850	-.137	.017	-.150	-.324	-.254	-.299	-.157	-.179	-.194	-.191
.900	-.114	.056	-.062	-.220	-.223	-.281	-.096	-.148	-.155	-.179

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(j) α = 0°; M = 0.92

z/b_v = 0.11 (Left side) and z/b_v = 0.38

$\frac{x}{c_v}$	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.000	.787	.699	.656	.452	.058	.625	.818	.680	.375	-.112
.025	.302	-.033	-.454	-.925	-.963	.348	-.054	-.930	-1.241	-.617
.075	.170	-.076	-.385	-.738	-.812	.172	-.119	-.672	-1.119	-.628
.150	.057	-.140	-.365	-.625	-.718	.076	-.136	-.391	-1.012	-.649
.250	.028	-.108	-.234	-.498	-.601	-.002	-.165	-.281	-.497	-.650
.350	-.034	-.143	-.212	-.265	-.528	-.096	-.225	-.319	-.126	-.605
.450	-.119	-.198	-.231	-.163	-.483	-.199	-.301	-.304	-.210	-.544
.550	-.150	-.214	-.184	-.155	-.414	-.250	-.361	-.291	-.233	-.483
.650	-.195	-.228	-.142	-.155	-.357	-.313	-.407	-.220	-.247	-.415
.750	-.202	-.157	-.063	-.131	-.291	-.313	-.267	-.127	-.228	-.357
.850	-.109	-.005	.015	-.072	-.226	-.209	.011	-.035	-.178	-.316
.900	-.039	.052	.052	-.037	-.186	-.078	.068	.014	-.130	-.284

Right side

$\frac{x}{c_v}$	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.025	-.322	.033	.350	.615	.845	-.885	-.026	.379	.603	.772
.075	-.404	-.083	.159	.369	.578	-.628	-.103	.175	.375	.558
.150	-.373	-.124	.063	.232	.411	-.362	-.127	.075	.245	.410
.250	-.247	-.113	.022	.156	.302	-.266	-.154	-.004	.134	.282
.350	-.219	-.152	-.047	.060	.187	-.295	-.214	-.089	.042	.176
.450	-.227	-.201	-.123	-.032	.082	-.280	-.272	-.171	-.072	.050
.550	-.181	-.226	-.160	-.083	.022	-.277	-.353	-.246	-.150	-.027
.650	-.124	-.214	-.172	-.117	-.028	-.202	-.393	-.295	-.202	-.090
.750	-.055	-.149	-.186	-.159	-.082	-.102	-.238	-.295	-.233	-.130
.850	.025	.000	-.073	-.105	-.073	-.017	.014	-.129	-.197	-.153
.900	.059	.052	-.023	-.084	-.098	.020	.061	-.058	-.184	-.198

z/b_v = 0.66 (Left side) and z/b_v = 0.93

$\frac{x}{c_v}$	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.000	.586	.822	.644	.316	-.130	.770	.852	.700	.514	.086
.025	.397	-.053	-1.031	-1.054	-.728	.258	-.535	-.784	-.351	-.393
.075	.220	-.109	-.840	-.946	-.740	-.039	-.520	-.335	-.341	-.389
.150	.086	-.164	-.312	-.674	-.760	-.244	-.594	-.335	-.340	-.392
.250	-.037	-.232	-.351	-.500	-.665	-.259	-.628	-.340	-.345	-.398
.350	-.172	-.361	-.277	-.302	-.471					
.450	-.312	-.489	-.360	-.334	-.431					
.550	-.391	-.583	-.356	-.357	-.437					
.650	-.484	-.602	-.246	-.376	-.455	-.772	-.275	-.317	-.371	-.423
.750	-.526	-.174	-.214	-.393	-.466	-.719	-.213	-.320	-.371	-.425
.850	-.481	.023	-.175	-.403	-.464	-.370	-.162	-.305	-.359	-.413
.900	-.248	.047	-.158	-.404	-.468	-.222	-.136	-.296	-.354	-.410

Right side

$\frac{x}{c_v}$	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.025	-.993	.000	.431	.637	.774	-.272	-.412	.330	.630	.986
.075	-.785	-.095	.212	.410	.586	-.307	-.521	-.065	.217	.480
.150	-.263	-.133	.102	.274	.443	-.316	-.563	-.180	.042	.260
.250	-.357	-.230	-.031	.131	.297					
.350	-.287	-.361	-.171	-.013	.150					
.450	-.344	-.482	-.287	-.147	.006					
.550	-.338	-.551	-.414	-.248	-.113					
.650	-.238	-.586	-.458	-.341	-.197	-.305	-.344	-.740	-.633	-.478
.750	-.213	-.267	-.521	-.430	-.287	-.314	-.293	-.638	-.658	-.520
.850	-.154	.020	-.315	-.391	-.307	-.303	-.201	-.266	-.555	-.386
.900	-.141	.066	-.134	-.418	-.349	-.284	-.143	-.199	-.397	-.262

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(k) α = 9.4°; M = 0.60

$\dfrac{x}{c_v}$	C_p for - z/b_v = 0.11					C_p for - z/b_v = 0.38				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
Left side										
.000	.414	.802	.435	-.710	-1.180	.151	.689	.110	-.733	-1.196
.025	-.015	-.136	-.264	-.719	-2.110	.352	-.111	-.814	-1.304	-.993
.075	.027	-.136	-.340	-.530	-1.021	.172	-.139	-.501	-1.240	-.968
.150	-.012	-.162	-.335	-.434	-.633	.075	-.148	-.374	-.922	-.920
.250	-.008	-.111	-.248	-.304	-.418	.011	-.152	-.319	-.361	-.872
.350	-.031	-.120	-.231	-.265	-.347	-.040	-.173	-.317	-.308	-.726
.450	-.049	-.143	-.229	-.235	-.340	-.077	-.192	-.298	-.290	-.573
.550	-.049	-.120	-.190	-.208	-.283	-.079	-.166	-.238	-.254	-.454
.650	-.029	-.086	-.137	-.176	-.215	-.065	-.129	-.183	-.219	-.379
.750	-.012	-.047	-.086	-.112	-.160	-.031	-.070	-.107	-.167	-.315
.850	.024	.002	-.017	-.037	-.089	.020	-.008	-.015	-.080	-.242
.900	.043	.043	.020	-.009	-.053	.047	.038	.031	-.037	-.192
Right side										
.025	-.195	-.044	.020	.523	.813	-.701	-.044	.409	.607	.719
.075	-.340	-.139	.031	.336	.548	-.469	-.113	.195	.393	.548
.150	-.324	-.143	-.001	.228	.393	-.365	-.120	.100	.256	.409
.250	-.254	-.107	-.006	.167	.290	-.303	-.134	.029	.162	.294
.350	-.241	-.118	-.031	.096	.192	-.289	-.152	-.017	.096	.210
.450	-.224	-.139	-.047	.034	.114	-.273	-.171	-.056	.032	.125
.550	-.190	-.116	-.042	.014	.080	-.231	-.159	-.070	.000	.078
.650	-.139	-.079	-.024	.016	.066	-.167	-.111	-.045	-.002	.057
.750	-.077	-.038	.001	.020	.052	-.091	-.042	-.012	.004	.039
.850	-.003	.031	.045	.055	.071	.004	.022	.043	.041	.041
.900	.031	.054	.064	.064	.064	.034	.043	.054	.036	.014

$\dfrac{x}{c_v}$	C_p for - z/b_v = 0.66					C_p for - z/b_v = 0.93				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
Left side										
.000	.006	.666	-.059	-.573	-.829	.296	.691	-.280	-.178	-.628
.025	.398	-.132	-1.128	-.975	-.692	.488	-.564	-1.188	-.464	-.450
.075	.216	-.150	-.621	-.909	-.715	.151	-.382	-1.107	-.450	-.441
.150	.105	-.175	-.469	-.722	-.639	-.036	-.359	-.987	-.445	-.445
.250	.024	-.192	-.413	-.514	-.582	-.077	-.320	-.757	-.454	-.454
.350	-.042	-.221	-.402	-.418	-.559					
.450	-.091	-.235	-.370	-.386	-.537					
.550	-.116	-.221	-.317	-.386	-.512					
.650	-.089	-.169	-.227	-.375	-.491	-.125	-.180	-.197	-.443	-.422
.750	-.045	-.093	-.132	-.349	-.468	-.063	-.086	-.114	-.418	-.411
.850	-.006	-.033	-.047	-.297	-.441	-.008	-.005	-.033	-.372	-.386
.900	.015	.002	-.008	-.258	-.432	.015	.031	.008	-.349	-.381
Right side										
.025	-.994	-.035	.451	.628	.694	-1.252	-.254	.580	.746	.950
.075	-.561	-.116	.234	.425	.571	-1.123	-.325	.165	.365	.600
.150	-.418	-.125	.130	.294	.447	-1.059	-.286	.017	.189	.393
.250	-.400	-.166	.045	.185	.336					
.350	-.393	-.203	-.024	.093	.237					
.450	-.358	-.219	-.072	.016	.144					
.550	-.298	-.203	-.093	-.025	.064					
.650	-.218	-.157	-.077	-.039	.018	-.169	-.129	-.072	-.082	-.043
.750	-.123	-.079	-.040	-.037	-.018	-.095	-.079	-.056	-.119	-.119
.850	-.019	.006	.029	-.018	-.043	-.003	.020	.024	-.105	-.135
.900	.027	.038	.052	-.032	-.089	.036	.052	.041	-.123	-.171

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(l) α = 9.6°; M = 0.80

$\frac{x}{c_v}$	C_p for -					C_p for -				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.370	.858	.414	-.395	-.170	.416	.737	.426	-.147	-.640
.025	-.032	-.111	-.154	-.580	-1.251	.341	-.093	-.968	-1.179	-.802
.075	.025	-.128	-.314	-.540	-.961	.166	-.136	-.493	-1.099	-.813
.150	-.006	-.163	-.354	-.483	-.733	.080	-.142	-.360	-.900	-.722
.250	-.007	-.107	-.248	-.329	-.493	.013	-.159	-.302	-.520	-.690
.350	-.038	-.125	-.233	-.279	-.402	-.053	-.199	-.299	-.333	-.633
.450	-.067	-.162	-.233	-.252	-.361	-.107	-.235	-.291	-.283	-.552
.550	-.067	-.142	-.199	-.218	-.316	-.107	-.206	-.234	-.255	-.480
.650	-.047	-.108	-.156	-.189	-.263	-.095	-.163	-.186	-.236	-.416
.750	-.024	-.061	-.091	-.145	-.201	-.053	-.082	-.114	-.196	-.364
.850	.017	.010	-.022	-.071	-.134	.013	.005	-.020	-.119	-.304
.900	.049	.050	.024	-.028	-.080	.051	.050	.032	-.068	-.249
Right side										
.025	-.095	-.016	.000	.415	.789	-1.055	-.039	.401	.611	.736
.075	-.333	-.130	.044	.331	.540	-.452	-.102	.192	.392	.561
.150	-.342	-.143	.020	.234	.395	-.357	-.120	.100	.266	.428
.250	-.259	-.105	-.002	.174	.296	-.305	-.137	.037	.169	.316
.350	-.253	-.127	-.036	.095	.196	-.303	-.171	-.022	.095	.228
.450	-.250	-.154	-.062	.023	.113	-.305	-.200	-.079	.018	.136
.550	-.207	-.136	-.059	-.002	.084	-.265	-.192	-.096	-.024	.081
.650	-.158	-.091	-.039	-.005	.058	-.191	-.143	-.080	-.034	.043
.750	-.087	-.052	-.019	-.007	.043	-.092	-.064	-.036	-.025	.028
.850	.000	.028	.041	.032	.055	.006	.027	.027	.012	.020
.900	.039	.056	.054	.035	.045	.043	.054	.041	-.001	-.017
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.298	.696	.306	-.085	-.610	.600	.757	.418	.129	-.380
.025	.391	-.105	-1.085	-.655	-.680	.420	-.620	-.771	-.390	-.392
.075	.221	-.130	-.636	-.685	-.745	.108	-.444	-.465	-.375	-.395
.150	.111	-.166	-.391	-.654	-.604	-.081	-.416	-.468	-.378	-.402
.250	.022	-.208	-.331	-.512	-.548	-.113	-.364	-.460	-.384	-.414
.350	-.063	-.269	-.333	-.410	-.507					
.450	-.138	-.304	-.336	-.372	-.486					
.550	-.176	-.286	-.306	-.381	-.472					
.650	-.142	-.203	-.240	-.390	-.460	-.184	-.166	-.314	-.410	-.410
.750	-.083	-.097	-.173	-.381	-.448	-.103	-.050	-.293	-.400	-.395
.850	-.017	-.009	-.096	-.350	-.423	-.049	.037	-.253	-.372	-.370
.900	.006	.034	-.060	-.329	-.411	-.027	.074	-.225	-.353	-.352
Right side										
.025	-1.038	-.016	.450	.628	.733	-.831	-.313	.495	.762	.995
.075	-.547	-.094	.244	.436	.605	-.708	-.384	.112	.374	.634
.150	-.379	-.114	.144	.314	.490	-.728	-.329	-.017	.194	.419
.250	-.383	-.179	.044	.207	.374					
.350	-.411	-.245	-.046	.092	.252					
.450	-.394	-.277	-.119	-.005	.149					
.550	-.325	-.266	-.154	-.061	.061					
.650	-.230	-.197	-.140	-.088	.001	-.231	-.140	-.143	-.139	-.066
.750	-.133	-.091	-.086	-.093	-.043	-.191	-.056	-.120	-.165	-.134
.850	-.021	.022	-.006	-.064	-.058	-.135	.050	-.057	-.138	-.145
.900	.022	.060	.018	-.078	-.099	-.095	.088	-.048	-.155	-.167

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(m) α = 9.7°; M = 0.85

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.349	.879	.420	-.236	-.046	.478	.748	.500	.009	-.513
.025	-.050	-.108	-.109	-.508	-1.154	.339	-.099	-1.172	-1.301	-.749
.075	.025	-.127	-.311	-.523	-.910	.167	-.144	-.441	-1.072	-.761
.150	-.006	-.171	-.381	-.525	-.723	.076	-.147	-.354	-.808	-.722
.250	-.013	-.117	-.256	-.341	-.536	.013	-.164	-.300	-.526	-.696
.350	-.049	-.138	-.237	-.281	-.457	-.059	-.221	-.300	-.345	-.651
.450	-.079	-.181	-.244	-.256	-.428	-.125	-.272	-.303	-.273	-.591
.550	-.076	-.165	-.207	-.216	-.370	-.132	-.249	-.254	-.241	-.526
.650	-.071	-.140	-.165	-.187	-.328	-.125	-.204	-.200	-.229	-.468
.750	-.043	-.084	-.105	-.153	-.260	-.079	-.111	-.126	-.199	-.409
.850	.010	-.005	-.027	-.083	-.179	-.005	-.002	-.031	-.136	-.341
.900	.043	.048	.022	-.042	-.135	.037	.051	.025	-.088	-.298
Right side										
.025	-.042	-.007	-.030	.344	.743	-1.010	-.032	.394	.615	.727
.075	-.320	-.127	.036	.307	.497	-.437	-.104	.189	.402	.542
.150	-.347	-.147	.018	.236	.352	-.346	-.121	.095	.274	.406
.250	-.254	-.111	-.004	.182	.265	-.294	-.141	.028	.179	.297
.350	-.248	-.133	-.046	.096	.173	-.295	-.184	-.031	.103	.206
.450	-.252	-.172	-.077	.024	.087	-.298	-.225	-.095	.014	.109
.550	-.211	-.161	-.077	-.006	.057	-.267	-.229	-.123	-.035	.054
.650	-.154	-.118	-.060	-.018	.024	-.191	-.181	-.109	-.053	.014
.750	-.093	-.067	-.037	-.022	.005	-.096	-.084	-.064	-.045	-.011
.850	-.005	.020	.029	.018	.014	.003	.025	.010	-.012	-.024
.900	.035	.052	.045	.028	-.004	.038	.054	.026	-.022	-.072

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.364	.715	.377	-.022	-.539	.669	.783	.502	.214	-.331
.025	.389	-.108	-1.208	-.676	-.690	.401	-.639	-.709	-.348	-.425
.075	.220	-.130	-.587	-.707	-.729	.091	-.474	-.391	-.337	-.425
.150	.109	-.170	-.381	-.669	-.687	-.106	-.447	-.391	-.343	-.431
.250	.018	-.225	-.321	-.515	-.713	-.125	-.362	-.394	-.351	-.445
.350	-.079	-.309	-.318	-.364	-.568					
.450	-.169	-.359	-.329	-.333	-.506					
.550	-.227	-.392	-.308	-.353	-.494					
.650	-.194	-.244	-.246	-.371	-.494	-.227	-.174	-.313	-.384	-.458
.750	-.119	-.110	-.187	-.380	-.491	-.142	-.056	-.308	-.384	-.444
.850	-.046	-.007	-.128	-.370	-.476	-.086	.038	-.286	-.365	-.421
.900	-.022	.038	-.095	-.355	-.467	-.072	.076	-.263	-.355	-.412
Right side										
.025	-1.189	.001	.446	.639	.724	-.599	-.313	.485	.770	.998
.075	-.463	-.087	.242	.448	.595	-.512	-.423	.102	.383	.621
.150	-.353	-.107	.146	.328	.478	-.522	-.343	-.027	.202	.401
.250	-.346	-.185	.039	.217	.361					
.350	-.369	-.268	-.063	.096	.234					
.450	-.374	-.329	-.148	-.008	.118					
.550	-.311	-.325	-.201	-.085	.022					
.650	-.234	-.239	-.191	-.120	-.049	-.280	-.138	-.190	-.179	-.134
.750	-.154	-.107	-.126	-.127	-.095	-.260	-.050	-.155	-.199	-.196
.850	-.058	.028	-.031	-.090	-.102	-.225	.062	-.086	-.162	-.192
.900	-.016	.072	-.011	-.109	-.151	-.194	.099	-.086	-.179	-.221

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TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = -6^0$ - Continued

(n) $\alpha = 9.7^0$; $M = 0.90$

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.306	.964	.440	-.126	.080	.534	.750	.563	.170	-.373
.025	-.079	-.042	.020	-.393	-.792	.339	-.046	-.976	-1.290	-.621
.075	.025	-.071	-.187	-.430	-.797	.170	-.097	-.448	-1.026	-.620
.150	.003	-.130	-.347	-.526	-.640	.080	-.109	-.373	-.738	-.626
.250	-.009	-.097	-.256	-.390	-.506	.011	-.135	-.279	-.497	-.617
.350	-.052	-.120	-.236	-.292	-.450	-.064	-.195	-.286	-.392	-.583
.450	-.091	-.183	-.257	-.281	-.424	-.145	-.271	-.322	-.307	-.535
.550	-.100	-.190	-.236	-.235	-.358	-.170	-.304	-.303	-.259	-.483
.650	-.104	-.204	-.206	-.209	-.319	-.190	-.309	-.243	-.244	-.434
.750	-.086	-.132	-.140	-.168	-.259	-.150	-.173	-.159	-.227	-.382
.850	-.022	-.026	-.060	-.109	-.175	-.051	-.013	-.062	-.172	-.325
.900	.018	.032	-.009	-.067	-.127	.007	.054	-.008	-.125	-.287
	Right side									
.025	.043	.055	-.027	.284	.760	-.954	.013	.400	.614	.754
.075	-.272	-.073	.046	.326	.516	-.532	-.065	.204	.405	.571
.150	-.395	-.113	.052	.276	.385	-.364	-.088	.114	.284	.436
.250	-.268	-.088	.033	.200	.295	-.276	-.116	.042	.184	.331
.350	-.255	-.120	-.014	.109	.204	-.295	-.173	-.025	.101	.237
.450	-.268	-.177	-.068	.018	.116	-.309	-.237	-.104	.005	.143
.550	-.236	-.199	-.091	-.021	.084	-.295	-.274	-.154	-.056	.075
.650	-.180	-.170	-.098	.045	.048	-.212	-.269	-.181	-.092	.033
.750	-.115	-.123	-.094	-.062	.024	-.114	-.158	-.163	-.104	.004
.850	-.024	-.013	-.032	-.028	.024	-.018	-.002	-.074	-.084	-.015
.900	.017	.028	-.005	-.024	.007	.019	.044	-.042	-.087	-.056
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.422	.721	.427	.080	-.452	.718	.815	.558	.293	-.240
.025	.394	-.053	-1.106	-.741	-.666	.404	-.479	-.754	-.378	-.416
.075	.230	-.084	-.911	-.746	-.704	.088	-.430	-.358	-.349	-.388
.150	.120	-.123	-.256	-.681	-.660	-.112	-.466	-.355	-.349	-.392
.250	.021	-.178	-.269	-.541	-.644	-.114	-.421	-.361	-.357	-.409
.350	-.084	-.299	-.304	-.344	-.508					
.450	-.196	-.396	-.351	-.342	-.459					
.550	-.298	-.443	-.339	-.367	-.453					
.650	-.310	-.445	-.265	-.381	-.454	-.574	-.244	-.317	-.385	-.419
.750	-.256	-.123	-.217	-.392	-.456	-.193	-.151	-.314	-.381	-.411
.850	-.100	.003	-.170	-.384	-.442	-.117	-.050	-.299	-.368	-.393
.900	-.059	.052	-.143	-.371	-.442	-.106	.007	-.284	-.355	-.382
	Right side									
.025	-1.121	.038	.453	.639	.760	-.393	-.274	.499	.769	1.012
.075	-.829	-.047	.261	.455	.623	-.391	-.441	.116	.381	.645
.150	-.271	-.073	.169	.336	.505	-.399	-.361	-.010	.200	.423
.250	-.301	-.160	.058	.225	.391					
.350	-.333	-.262	-.052	.099	.266					
.450	-.375	-.336	-.162	-.017	.153					
.550	-.324	-.408	-.240	-.118	.046					
.650	-.243	-.416	-.312	-.179	-.029	-.303	-.254	-.566	-.441	-.132
.750	-.188	-.206	-.322	-.250	-.086	-.301	-.156	-.431	-.417	-.188
.850	-.104	.029	-.170	-.205	-.089	-.281	-.029	-.136	-.234	-.179
.900	-.068	.074	-.093	-.185	-.131	-.252	.038	-.115	-.205	-.208

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(o) α = 9.7°; M = 0.92

$\frac{x}{c_v}$	C_p for –					C_p for –				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.312	.971	.453	-.173		.569	.761	.588	.176	
.025	-.042	-.025	.044	-.371		.356	-.029	-.959	-1.302	
.075	.058	-.054	-.166	-.415		.190	-.079	-.402	-1.039	
.150	.045	-.106	-.325	-.516		.097	-.097	-.337	-.737	
.250	.021	-.068	-.227	-.391		.025	-.121	-.271	-.502	
.350	-.024	-.106	-.219	-.297		-.050	-.182	-.287	-.390	
.450	-.073	-.168	-.245	-.279		-.138	-.258	-.317	-.301	
.550	-.094	-.177	-.224	-.238		-.167	-.296	-.291	-.256	
.650	-.105	-.188	-.194	-.211		-.200	-.291	-.224	-.248	
.750	-.094	-.121	-.127	-.172		-.175	-.167	-.139	-.229	
.850	-.033	-.016	-.042	-.115		-.067	.001	-.042	-.175	
.900	.010	.037	.003	-.072		-.007	.056	.015	-.130	
Right side										
.025	.124	.069	.003	.282		-.855	.023	.413	.614	
.075	-.178	-.060	.073	.333		-.430	-.054	.217	.405	
.150	-.307	-.095	.068	.285		-.323	-.077	.120	.281	
.250	-.225	-.071	.039	.199		-.257	-.108	.054	.180	
.350	-.235	-.109	-.010	.106		-.285	-.157	-.017	.095	
.450	-.257	-.164	-.064	.019		-.300	-.226	-.094	.002	
.550	-.237	-.181	-.086	-.024		-.306	-.262	-.144	-.060	
.650	-.183	-.160	-.089	-.053		-.227	-.260	-.162	-.102	
.750	-.120	-.117	-.081	-.076		-.117	-.144	-.137	-.118	
.850	-.020	.001	-.017	-.039		-.017	.012	-.045	-.093	
.900	.018	.033	.010	-.035		.022	.050	-.015	-.101	

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
Left side										
.000	.443	.731	.466	.088		.721	.822	.583	.303	
.025	.399	-.039	-1.076	-.742		.399	-.456	-.768	-.382	
.075	.240	-.070	-.847	-.740		.091	-.416	-.349	-.354	
.150	.133	-.110	-.264	-.682		-.107	-.454	-.350	-.358	
.250	.028	-.171	-.268	-.569		-.102	-.413	-.357	-.358	
.350	-.078	-.290	-.295	-.335						
.450	-.183	-.388	-.341	-.338						
.550	-.293	-.437	-.317	-.363						
.650	-.345	-.435	-.250	-.382		-.619	-.223	-.299	-.387	
.750	-.307	-.116	-.194	-.396		-.315	-.138	-.300	-.390	
.850	-.129	.012	-.147	-.392		-.117	-.042	-.287	-.375	
.900	-.067	.052	-.116	-.380		-.100	.001	-.266	-.367	
Right side										
.025	-1.049	.042	.460	.641		-.436	-.270	.488	.765	
.075	-.559	-.038	.267	.448		-.389	-.435	.112	.373	
.150	-.265	-.066	.167	.333		-.395	-.355	-.017	.190	
.250	-.310	-.152	.065	.223						
.350	-.319	-.260	-.048	.080						
.450	-.381	-.332	-.158	-.025						
.550	-.348	-.405	-.236	-.128						
.650	-.233	-.412	-.300	-.190		-.291	-.268	-.555	-.456	
.750	-.181	-.203	-.290	-.263		-.291	-.155	-.323	-.478	
.850	-.099	.033	-.118	-.233		-.266	-.022	-.106	-.267	
.900	-.063	.072	-.051	-.217		-.242	.035	-.091	-.222	

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = -6°$ - Continued

(p) $\alpha = 15.6°$; $M = 0.60$

$\frac{x}{c_v}$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	-.351	.825	.182	.091	-.489	.262	.740	.364	-.443	-.879
.025	-.619	-.139	.278	.098	-.491	.157	-.102	-.347	-.694	-.931
.075	-.171	-.143	.014	-.091	-.417	.075	-.143	-.309	-.507	-.798
.150	-.045	-.171	-.143	-.199	-.394	.055	-.139	-.282	-.391	-.647
.250	.007	-.127	-.152	-.180	-.331	.034	-.153	-.270	-.340	-.532
.350	.027	-.112	-.179	-.199	-.293	.011	-.182	-.275	-.315	-.451
.450	.027	-.127	-.197	-.203	-.261	-.009	-.203	-.270	-.288	-.383
.550	.025	-.121	-.170	-.176	-.213	.000	-.171	-.225	-.244	-.326
.650	.034	-.086	-.132	-.135	-.168	.000	-.139	-.177	-.206	-.279
.750	.034	-.057	-.093	-.096	-.123	.009	-.089	-.116	-.149	-.218
.850	.034	-.009	-.036	-.041	-.067	.034	-.018	-.043	-.087	-.150
.900	.052	.027	-.004	-.009	-.035	.043	.023	-.002	-.046	-.107
	Right side									
.025	.344	-.098	-.732	-.260	-.089	-.460	-.096	.123	.196	.211
.075	-.023	-.175	-.207	-.075	.030	-.364	-.139	.050	.123	.245
.150	-.162	-.164	-.059	.004	.112	-.317	-.148	.034	.107	.256
.250	-.175	-.125	.005	.055	.161	-.285	-.150	.032	.109	.270
.350	-.200	-.121	.025	.075	.182	-.278	-.166	.021	.100	.261
.450	-.205	-.137	.023	.080	.179	-.266	-.182	.003	.087	.233
.550	-.171	-.118	.028	.080	.159	-.230	-.168	-.004	.078	.204
.650	-.130	-.082	.032	.075	.132	-.173	-.130	.009	.073	.170
.750	-.086	-.052	.037	.068	.109	-.107	-.073	.028	.071	.143
.850	-.018	.011	.057	.073	.093	-.034	.002	.053	.075	.116
.900	.009	.032	.053	.066	.075	-.004	.018	.046	.059	.084
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	-.043	.665	.062	-.530	-.760	.248	.731	-.347	-.496	-.708
.025	.453	-.100	-1.027	-1.103	-.744	.627	-.399	-1.109	-.612	-.491
.075	.276	-.121	-.577	-.998	-.724	.285	-.305	-1.100	-.601	-.462
.150	.171	-.148	-.438	-.806	-.681	.100	-.287	-.968	-.619	-.475
.250	.103	-.166	-.384	-.569	-.631	.059	-.228	-.707	-.598	-.466
.350	.043	-.203	-.372	-.452	-.579					
.450	-.002	-.221	-.354	-.395	-.520					
.550	-.036	-.212	-.297	-.352	-.460					
.650	-.023	-.162	-.218	-.288	-.387	-.082	-.178	-.188	-.386	-.329
.750	.000	-.091	-.132	-.212	-.331	-.027	-.082	-.107	-.322	-.295
.850	.021	-.032	-.054	-.137	-.272	.007	-.002	-.027	-.256	-.272
.900	.027	.000	-.009	-.112	-.243	.021	.032	.016	-.231	-.250
	Right side									
.025	-1.105	-.064	.455	.621	.610	-1.086	-.287	.628	.822	.960
.075	-.642	-.121	.255	.422	.502	-1.052	-.278	.264	.470	.656
.150	-.430	-.123	.164	.310	.428	-.900	-.262	.103	.278	.457
.250	-.392	-.164	.100	.228	.362					
.350	-.387	-.196	.037	.148	.297					
.450	-.351	-.209	.000	.105	.240					
.550	-.296	-.203	-.022	.068	.184					
.650	-.219	-.162	-.013	.052	.148	-.168	-.091	-.002	.050	.098
.750	-.134	-.089	.009	.048	.103	-.116	-.073	-.009	-.009	.001
.850	-.032	.000	.053	.062	.078	-.032	.018	.048	.000	-.024
.900	.009	.030	.068	.055	.035	.005	.052	.062	-.016	-.053

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(q) α = 15.8°; M = 0.80

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.032	.859	.404	.126	-.499	.332	.778	.299	-.229	-.555
.025	-.955	.095	.327	.051	-.452	-.002	-.049	-.314	-.534	-.645
.075	-.574	-.097	.007	-.121	-.410	-.067	-.139	-.323	-.438	-.584
.150	-.260	-.197	-.192	-.236	-.386	-.059	-.156	-.300	-.380	-.517
.250	-.124	-.142	-.179	-.197	-.325	-.041	-.162	-.286	-.339	-.463
.350	-.061	-.123	-.197	-.204	-.278	-.041	-.197	-.289	-.318	-.413
.450	-.017	-.139	-.215	-.207	-.234	-.041	-.227	-.279	-.294	-.360
.550	.001	-.132	-.188	-.171	-.189	-.018	-.201	-.242	-.254	-.311
.650	.024	-.091	-.144	-.138	-.152	-.008	-.157	-.197	-.223	-.269
.750	.035	-.055	-.094	-.103	-.116	.009	-.095	-.127	-.168	-.214
.850	.047	-.003	-.040	-.046	-.066	.041	-.009	-.051	-.106	-.157
.900	.057	.033	-.007	-.016	-.033	.054	.036	-.009	-.066	-.122
	Right side									
.025	.394	-.768	-.948	-.452	-.302	-.420	-.269	-.050	.005	.055
.075	-.014	-.413	-.585	-.250	-.172	-.360	-.260	-.092	-.028	.102
.150	-.210	-.251	-.279	-.166	-.048	-.320	-.201	-.072	-.016	.164
.250	-.204	-.139	-.136	-.081	.058	-.292	-.154	-.040	.017	.220
.350	-.221	-.088	-.065	-.019	.130	-.283	-.142	-.030	.034	.242
.450	-.222	-.077	-.022	.019	.163	-.283	-.151	-.027	.045	.236
.550	-.191	-.053	.008	.043	.157	-.249	-.138	-.013	.054	.217
.650	-.141	-.021	.026	.055	.139	-.189	-.094	.007	.064	.193
.750	-.095	-.002	.037	.061	.117	-.115	-.036	.031	.067	.157
.850	-.021	.042	.057	.073	.093	-.035	.030	.061	.076	.127
.900	.000	.053	.060	.064	.078	-.003	.047	.052	.061	.090
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.364	.708	.454	.070	-.390	.625	.790	.356	-.075	-.523
.025	.380	-.061	-1.059	-.918	-.631	.529	-.497	-.934	-.508	-.499
.075	.220	-.110	-.532	-.810	-.611	.202	-.363	-.582	-.406	-.401
.150	.128	-.147	-.402	-.649	-.584	.038	-.345	-.603	-.408	-.411
.250	.066	-.186	-.347	-.468	-.542	.024	-.248	-.573	-.408	-.408
.350	.009	-.237	-.356	-.373	-.505					
.450	-.046	-.277	-.361	-.330	-.463					
.550	-.071	-.266	-.315	-.309	-.414					
.650	-.053	-.198	-.232	-.283	-.363	-.129	-.169	-.277	-.374	-.284
.750	-.018	-.097	-.151	-.251	-.314	-.059	-.047	-.239	-.348	-.257
.850	.016	-.014	-.068	-.201	-.270	-.021	.042	-.186	-.311	-.230
.900	.030	.029	-.031	-.171	-.248	-.015	.075	-.153	-.292	-.220
	Right side									
.025	-1.112	-.085	.400	.597	.687	-.624	-.289	.542	.799	1.013
.075	-.536	-.126	.196	.381	.557	-.610	-.302	.193	.435	.687
.150	-.373	-.109	.137	.286	.480	-.632	-.260	.051	.246	.486
.250	-.357	-.144	.072	.205	.413					
.350	-.378	-.186	.005	.125	.342					
.450	-.369	-.204	-.037	.076	.278					
.550	-.310	-.197	-.056	.049	.217					
.650	-.230	-.136	-.040	.040	.174	-.246	-.077	-.021	.028	.105
.750	-.147	-.059	-.009	.037	.128	-.225	-.024	-.039	-.034	.004
.850	-.042	.039	.048	.051	.084	-.159	.071	.010	-.025	-.022
.900	.004	.074	.061	.037	.048	-.126	.107	.010	-.050	-.054

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Continued

(r) α = 15.9°; M = 0.85

$z/b_v = 0.11$ (Left group) and $z/b_v = 0.38$ (Right group)

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
					Left side					
.000	.309	.904	.574	.140	-.587	.420	.775	.349	-.237	-.522
.025	-1.069	.002	.247	-.063	-.552	.097	-.153	-.435	-.595	-.646
.075	-.603	-.205	-.046	-.211	-.498	-.041	-.230	-.418	-.504	-.594
.150	-.288	-.298	-.278	-.320	-.457	-.072	-.216	-.368	-.438	-.533
.250	-.179	-.206	-.233	-.254	-.387	-.079	-.204	-.326	-.382	-.475
.350	-.130	-.164	-.235	-.240	-.322	-.098	-.235	-.318	-.348	-.432
.450	-.092	-.162	-.241	-.233	-.265	-.113	-.273	-.309	-.321	-.384
.550	-.055	-.162	-.219	-.197	-.209	-.085	-.242	-.274	-.277	-.336
.650	-.025	-.104	-.159	-.149	-.168	-.062	-.192	-.220	-.241	-.299
.750	.000	-.060	-.111	-.117	-.130	-.034	-.109	-.151	-.187	-.230
.850	.023	-.009	-.052	-.063	-.076	.006	-.018	-.069	-.126	-.188
.900	.046	.027	-.013	-.035	-.044	.030	.033	-.023	-.083	-.148
					Right side					
.025	.318	-.685	-1.070	-.511	-.418	-.791	-.279	-.069	-.052	-.016
.075	-.096	-.845	-.664	-.324	-.285	-.573	-.315	-.146	-.096	.014
.150	-.354	-.330	-.337	-.241	-.152	-.427	-.262	-.145	-.086	.079
.250	-.284	-.233	-.220	-.166	-.034	-.312	-.211	-.108	-.052	.142
.350	-.254	-.160	-.153	-.092	.058	-.298	-.194	-.094	-.032	.169
.450	-.238	-.126	-.094	-.038	.111	-.295	-.201	-.084	-.013	.183
.550	-.190	-.073	-.047	-.004	.124	-.266	-.178	-.066	.002	.178
.650	-.150	-.042	-.012	.021	.114	-.207	-.124	-.036	.026	.163
.750	-.099	-.015	.013	.036	.104	-.127	-.058	-.004	.039	.134
.850	-.021	.030	.043	.052	.086	-.047	.023	.036	.052	.110
.900	.002	.049	.045	.051	.066	-.011	.040	.033	.038	.068

$z/b_v = 0.66$ (Left group) and $z/b_v = 0.93$ (Right group)

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
					Left side					
.000	.368	.709	.496	.166	-.234	.680	.809	.497	.061	-.420
.025	.368	-.102	-1.170	-.941	-.621	.498	-.555	-.886	-.448	-.512
.075	.200	-.144	-.544	-.834	-.605	.172	-.422	-.469	-.374	-.402
.150	.105	-.175	-.400	-.662	-.578	-.011	-.386	-.484	-.375	-.411
.250	.037	-.215	-.336	-.489	-.540	-.017	-.239	-.479	-.380	-.415
.350	-.040	-.279	-.346	-.375	-.501					
.450	-.113	-.334	-.359	-.324	-.468					
.550	-.147	-.347	-.318	-.303	-.428					
.650	-.120	-.230	-.247	-.287	-.384	-.201	-.165	-.313	-.377	-.320
.750	-.074	-.111	-.175	-.271	-.343	-.116	-.049	-.295	-.360	-.291
.850	-.031	-.017	-.104	-.236	-.299	-.072	.040	-.257	-.337	-.257
.900	-.015	.025	-.064	-.216	-.282	-.069	.074	-.233	-.327	-.248
					Right side					
.025	-1.301	-.099	.359	.576	.708	-.498	-.290	.487	.780	1.006
.075	-.710	-.157	.162	.357	.551	-.486	-.381	.134	.398	.673
.150	-.320	-.143	.095	.254	.464	-.496	-.320	-.015	.204	.455
.250	-.326	-.187	.028	.170	.389					
.350	-.349	-.236	-.049	.086	.310					
.450	-.359	-.256	-.094	.039	.247					
.550	-.316	-.249	-.114	-.001	.193					
.650	-.249	-.177	-.098	-.005	.145	-.301	-.077	-.059	-.018	.094
.750	-.180	-.079	-.054	-.002	.103	-.289	-.029	-.083	-.073	-.028
.850	-.085	.034	.016	.021	.068	-.245	.074	-.030	-.056	-.051
.900	-.041	.070	.029	.001	.024	-.208	.105	-.037	-.083	-.095

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = -6° - Concluded

(s) α = 15.9°; M = 0.90

$\frac{x}{c_v}$	C_p for - $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for - $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.560	.906	.777	.280		.462	.756	.517	.166	
.025	-1.029	-.264	.238	-.202		.349	-.138	-1.117	-1.363	
.075	-.512	-.365	-.063	-.295		.150	-.220	-.593	-.926	
.150	-.178	-.368	-.354	-.480		.049	-.240	-.567	-.754	
.250	-.138	-.339	-.605	-.658		-.027	-.226	-.607	-.698	
.350	-.145	-.246	-.497	-.584		-.110	-.269	-.295	-.420	
.450	-.161	-.234	-.229	-.280		-.186	-.344	-.293	-.404	
.550	-.209	-.324	-.319	-.324		-.212	-.388	-.346	-.353	
.650	-.197	-.229	-.231	-.268		-.230	-.340	-.286	-.297	
.750	-.164	-.144	-.163	-.214		-.198	-.198	-.191	-.244	
.850	-.105	-.054	-.083	-.146		-.120	-.050	-.101	-.188	
.900	-.060	-.003	-.030	-.092		-.176	-.194	-.175	-.154	
Right side										
.025	.330	-.226	-.951	-.105		-1.101	-.130	.371	.461	
.075	-.073	-.487	-.489	-.154		-.618	-.214	.145	.236	
.150	-.377	-.360	-.182	-.140		-.598	-.230	.039	.111	
.250	-.619	-.320	-.150	-.137		-.596	-.218	-.029	.035	
.350	-.491	-.261	-.165	-.136		-.319	-.265	-.091	-.032	
.450	-.273	-.289	-.189	-.149		-.284	-.319	-.166	-.101	
.550	-.241	-.241	-.189	-.118		-.332	-.333	-.203	-.130	
.650	-.218	-.205	-.186	-.112		-.284	-.320	-.213	-.140	
.750	-.172	-.150	-.158	-.088		-.188	-.198	-.183	-.124	
.850	-.072	-.040	-.073	-.044		-.090	-.042	-.093	-.081	
.900	-.038	-.006	-.047	-.038		-.052	-.006	-.065	-.096	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.321	.695	.366	-.015		.699	.823	.499	.079	
.025	.409	-.088	-1.231	-.746		.535	-.505	-.821	-.380	
.075	.244	-.124	-1.042	-.747		.201	-.463	-.391	-.373	
.150	.131	-.160	-.622	-.748		-.006	-.391	-.393	-.375	
.250	.039	-.214	-.207	-.663		-.015	-.258	-.405	-.394	
.350	-.070	-.323	-.270	-.547						
.450	-.190	-.415	-.375	-.465						
.550	-.300	-.467	-.401	-.369						
.650	-.312	-.504	-.318	-.351		-.546	-.212	-.367	-.423	
.750	-.324	-.144	-.267	-.365		-.469	-.165	-.365	-.427	
.850	-.230	-.034	-.214	-.369		-.196	-.096	-.342	-.420	
.900	-.168	.003	-.179	-.360		-.174	-.050	-.321	-.415	
Right side										
.025	-1.258	-.049	.431	.600		-.421	-.216	.547	.823	
.075	-1.046	-.116	.233	.412		-.401	-.459	.179	.437	
.150	-.574	-.125	.143	.298		-.403	-.345	.009	.217	
.250	-.198	-.206	.042	.194						
.350	-.289	-.311	-.069	.062						
.450	-.373	-.377	-.174	-.050						
.550	-.387	-.440	-.262	-.145						
.650	-.319	-.439	-.323	-.200		-.363	-.309	-.491	-.293	
.750	-.276	-.288	-.338	-.222		-.369	-.218	-.334	-.269	
.850	-.205	-.015	-.153	-.152		-.349	-.081	-.138	-.202	
.900	-.176	.029	-.101	-.166		-.319	-.018	-.135	-.226	

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$

(a) $\alpha = 0°$; M = 0.60

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.642	.647	.334	-.438	-1.286	.262	.741	.289	-.688	-1.166
.025	.276	-.096	-.561	-1.134	-2.145	.346	-.098	-.647	-1.405	-1.064
.075	.141	-.105	-.372	-.672	-1.504	.160	-.131	-.443	-1.063	-1.038
.150	.048	-.133	-.307	-.500	-.677	.066	-.141	-.329	-.573	-.967
.250	.023	-.100	-.213	-.307	-.404	.002	-.138	-.279	-.403	-.907
.350	-.011	-.112	-.182	-.240	-.325	-.055	-.176	-.272	-.348	-.772
.450	-.052	-.124	-.168	-.201	-.284	-.089	-.190	-.252	-.297	-.613
.550	-.057	-.105	-.127	-.157	-.284	-.082	-.150	-.200	-.231	-.478
.650	-.039	-.074	-.088	-.113	-.219	-.066	-.115	-.145	-.171	-.383
.750	-.014	-.029	-.034	-.072	-.140	-.030	-.065	-.079	-.100	-.293
.850	.030	.025	.023	-.015	-.069	.027	.006	-.002	-.021	-.200
.900	.055	.051	.055	.020	-.022	.052	.041	.041	.025	-.150
Right side										
.025	-.449	-.034	.309	.576	.797	-.645	-.065	.382	.608	.749
.075	-.389	-.115	.141	.349	.550	-.435	-.126	.171	.379	.533
.150	-.305	-.122	.062	.220	.383	-.337	-.133	.078	.238	.378
.250	-.219	-.105	.030	.149	.269	-.273	-.143	.016	.142	.246
.350	-.194	-.115	-.018	.077	.175	-.253	-.157	-.032	.071	.168
.450	-.173	-.129	-.057	.022	.100	-.241	-.162	-.070	.022	.098
.550	-.132	-.105	-.054	.006	.068	-.203	-.145	-.072	-.003	.054
.650	-.080	-.065	-.025	.015	.059	-.143	-.107	-.047	.008	.043
.750	-.030	-.027	-.002	.031	.056	-.071	-.046	-.011	.029	.038
.850	.034	.034	.050	.061	.073	.007	.030	.055	.064	.047
.900	.057	.053	.064	.071	.073	.039	.041	.062	.064	.022
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.212	.732	.239	-.451	-.640	.358	.772	.114	-.490	-.610
.025	.374	-.115	-.875	-1.146	-.691	.483	-.058	-1.002	-1.015	-.545
.075	.185	-.138	-.513	-1.086	-.668	.330	-.070	-.572	-.842	-.494
.150	.082	-.155	-.400	-.975	-.670	.180	-.129	-.475	-.787	-.497
.250	.016	-.157	-.332	-.709	-.629	.052	-.195	-.422	-.677	-.497
.350	-.036	-.181	-.311	-.467	-.580					
.450	-.080	-.200	-.297	-.359	-.545					
.550	-.102	-.200	-.268	-.302	-.506					
.650	-.080	-.157	-.209	-.238	-.464	-.143	-.235	-.293	-.346	-.381
.750	-.041	-.105	-.134	-.164	-.423	-.118	-.181	-.216	-.288	-.353
.850	-.007	-.037	-.057	-.097	-.381	-.080	-.112	.129	-.226	-.316
.900	.009	-.006	-.016	-.061	-.363	-.050	-.074	-.086	-.189	-.300
Right side										
.025	-.802	-.070	.414	.615	.702	-.779	.013	.514	.675	.735
.075	-.506	-.129	.191	.395	.519	-.526	-.051	.323	.528	.658
.150	-.362	-.138	.100	.259	.385	-.433	-.110	.203	.404	.561
.250	-.326	-.155	.028	.165	.274					
.350	-.314	-.176	-.029	.091	.188					
.450	-.296	-.190	-.061	.038	.124					
.550	-.262	-.183	-.077	.011	.070					
.650	-.205	-.152	-.066	.002	.033	-.273	-.195	-.102	-.033	-.011
.750	-.134	-.110	-.032	.008	.001	-.207	-.171	-.102	-.072	-.092
.850	-.032	.011	.025	.043	-.022	-.100	-.065	-.032	-.042	-.108
.900	.011	.027	.050	.043	-.069	-.055	-.037	-.009	-.051	-.140

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$ - Continued

(b) $\alpha = 0°$; $M = 0.80$

$\dfrac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.725	.677	.523	.201	-.240	.475	.775	.520	.010	-.428
.025	.286	-.069	-.548	-1.042	-1.000	.347	-.069	-.983	-1.715	-.683
.075	.155	-.090	-.374	-.872	-.884	.163	-.122	-.462	-1.506	-.688
.150	.055	-.133	-.330	-.549	-.798	.071	-.141	-.352	-.482	-.702
.250	.029	-.097	-.228	-.312	-.664	.000	-.152	-.301	-.370	-.686
.350	-.015	-.111	-.196	-.246	-.539	-.061	-.174	-.290	-.349	-.616
.450	-.063	-.133	-.187	-.218	-.452	-.110	-.197	-.279	-.303	-.531
.550	-.063	-.111	-.115	-.162	-.341	-.096	-.169	-.225	-.234	-.438
.650	-.054	-.082	-.095	-.124	-.266	-.078	-.127	-.164	-.174	-.359
.750	-.020	-.027	-.030	-.071	-.199	-.032	-.060	-.085	-.096	-.297
.850	.032	.033	.035	-.009	-.116	.030	.019	.008	-.009	-.250
.900	.059	.069	.064	.031	-.064	.065	.061	.055	.037	-.213
Right side										
.025	-.400	.004	.324	.593	.810	-.977	-.057	.379	.608	.752
.075	-.395	-.094	.149	.360	.560	-.442	-.115	.166	.374	.545
.150	-.313	-.115	.066	.230	.394	-.343	-.127	.076	.239	.394
.250	-.229	-.099	.028	.159	.294	-.287	-.139	.006	.144	.276
.350	-.203	-.118	-.023	.079	.190	-.272	-.161	-.042	.074	.187
.450	-.191	-.129	-.061	.016	.104	-.265	-.175	-.083	.010	.104
.550	-.144	-.113	-.064	-.003	.064	-.223	-.158	-.091	-.016	.055
.650	-.084	-.066	-.036	.005	.044	-.156	-.111	-.061	-.009	.038
.750	-.029	-.027	-.009	.019	.037	-.069	-.041	-.012	.013	.023
.850	.042	.046	.047	.059	.047	.023	.043	.052	.057	.017
.900	.074	.065	.060	.066	.040	.059	.058	.060	.056	-.023
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.440	.764	.488	-.003	-.336	.533	.811	.397	-.139	-.349
.025	.370	-.099	-1.152	-1.336	-.735	.483	-.012	-1.167	-1.093	-.402
.075	.192	-.129	-.565	-1.182	-.761	.350	-.044	-.691	-.577	-.390
.150	.081	-.150	-.422	-.950	-.773	.203	-.113	-.545	-.580	-.414
.250	.018	-.157	-.345	-.692	-.689	.065	-.202	-.425	-.589	-.423
.350	-.037	-.186	-.334	-.480	-.484					
.450	-.087	-.220	-.340	-.373	-.449					
.550	-.118	-.224	-.314	-.332	-.451					
.650	-.096	-.177	-.238	-.283	-.455	-.163	-.253	-.308	-.382	-.399
.750	-.054	-.107	-.143	-.215	-.447	-.137	-.182	-.208	-.318	-.384
.850	-.005	-.027	-.047	-.142	-.414	-.081	-.097	-.106	-.258	-.344
.900	.023	.007	-.003	-.102	-.400	-.043	-.049	-.059	-.235	-.317
Right side										
.025	-1.220	-.054	.408	.624	.731	-1.160	.055	.525	.705	.782
.075	-.517	-.125	.187	.396	.545	-.601	-.024	.354	.559	.706
.150	-.371	-.122	.099	.272	.411	-.476	-.074	.230	.435	.607
.250	-.333	-.146	.025	.172	.300					
.350	-.331	-.171	-.030	.097	.212					
.450	-.330	-.206	-.068	.040	.145					
.550	-.299	-.206	-.092	.001	.078					
.650	-.226	-.171	-.079	-.016	.034	-.297	-.227	-.123	-.067	-.018
.750	-.131	-.111	-.042	-.016	-.012	-.197	-.171	-.115	-.117	-.119
.850	-.017	.023	.032	.019	-.044	-.072	-.051	-.026	-.085	-.125
.900	.035	.044	.055	.022	-.090	-.020	-.016	-.003	-.099	-.161

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$ - Continued

(c) $\alpha = 0°$; M = 0.85

	C_p for -					C_p for -				
$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.757	.682	.574	.327	-.098	.531	.789	.611	.166	-.292
.025	.283	-.061	-.522	-.946	-.975	.342	-.072	-1.052	-1.453	-.665
.075	.156	-.091	-.369	-.828	-.849	.160	-.131	-.424	-1.287	-.667
.150	.054	-.138	-.343	-.648	-.751	.070	-.145	-.355	-1.113	-.678
.250	.029	-.111	-.232	-.343	-.626	-.004	-.159	-.306	-.276	-.658
.350	-.020	-.122	-.202	-.241	-.535	-.066	-.189	-.302	-.287	-.607
.450	-.070	-.148	-.199	-.206	-.478	-.122	-.218	-.292	-.296	-.535
.550	-.073	-.128	-.152	-.160	-.387	-.109	-.194	-.239	-.240	-.455
.650	-.064	-.102	-.106	-.131	-.311	-.087	-.152	-.177	-.181	-.381
.750	-.027	-.045	-.034	-.077	-.240	-.040	-.078	-.087	-.110	-.306
.850	.026	.029	.036	-.013	-.153	.029	.015	.013	-.020	-.260
.900	.057	.068	.067	.029	-.108	.070	.065	.061	.033	-.224
Right side										
.025	-.369	.010	.329	.596	.814	-.967	-.045	.373	.599	.744
.075	-.388	-.092	.147	.360	.555	-.441	-.118	.166	.369	.536
.150	-.322	-.116	.066	.230	.387	-.349	-.134	.074	.236	.383
.250	-.238	-.108	.027	.159	.281	-.295	-.152	.004	.137	.261
.350	-.216	-.129	-.027	.074	.181	-.282	-.175	-.052	.059	.176
.450	-.202	-.148	-.073	.003	.087	-.275	-.194	-.097	-.007	.087
.550	-.156	-.132	-.077	-.018	.047	-.236	-.181	-.104	-.036	.034
.650	-.090	-.079	-.046	-.010	.024	-.170	-.135	-.073	-.028	.014
.750	-.032	-.038	-.022	-.001	.011	-.072	-.059	-.022	-.006	-.002
.850	.043	.038	.046	.049	.024	.026	.042	.050	.040	-.012
.900	.074	.070	.064	.059	.008	.063	.062	.061	.040	-.057
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.498	.777	.554	.147	-.237	.569	.819	.458	-.021	-.290
.025	.363	-.102	-1.167	-1.411	-.743	.488	-.048	-1.153	-1.153	-.463
.075	.183	-.141	-.575	-1.214	-.762	.358	-.051	-.787	-.511	-.392
.150	.079	-.164	-.412	-.901	-.772	.205	-.114	-.579	-.508	-.415
.250	.011	-.168	-.333	-.638	-.713	.060	-.216	-.411	-.513	-.431
.350	-.046	-.205	-.341	-.450	-.477					
.450	-.093	-.245	-.358	-.346	-.439					
.550	-.133	-.259	-.332	-.324	-.442					
.650	-.113	-.213	-.258	-.298	-.451	-.187	-.289	-.311	-.413	-.408
.750	-.067	-.128	-.142	-.251	-.454	-.150	-.199	-.203	-.370	-.399
.850	-.009	-.028	-.034	-.191	-.435	-.084	-.098	-.094	-.313	-.367
.900	.020	.013	.010	-.150	-.418	-.044	-.047	-.043	-.274	-.350
Right side										
.025	-1.150	-.047	.403	.610	.727	-1.206	.069	.524	.703	.792
.075	-.435	-.124	.190	.387	.537	-.630	-.019	.353	.563	.707
.150	-.365	-.126	.097	.263	.399	-.482	-.077	.230	.432	.603
.250	-.329	-.162	.024	.163	.289					
.350	-.342	-.195	-.039	.089	.200					
.450	-.353	-.226	-.086	.029	.122					
.550	-.343	-.241	-.109	-.023	.057					
.650	-.253	-.202	-.097	-.043	.002	-.306	-.251	-.136	-.098	-.056
.750	-.140	-.125	-.057	-.047	-.047	-.196	-.182	-.132	-.156	-.156
.850	-.013	.012	.024	-.014	-.067	-.062	-.051	-.037	-.123	-.159
.900	.040	.045	.053	-.014	-.122	-.012	-.011	-.009	-.133	-.199

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$ - Continued

(d) $\alpha = 0°$; M = 0.90

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.786	.693	.619	.436	.052	.585	.798	.670	.300	-.158
.025	.297	-.050	-.472	-.861	-.844	.345	-.069	-.935	-1.268	-.607
.075	.169	-.084	-.377	-.747	-.732	.165	-.138	-.595	-1.141	-.614
.150	.063	-.144	-.362	-.648	-.688	.073	-.155	-.393	-1.036	-.632
.250	.033	-.117	-.252	-.489	-.610	.002	-.172	-.324	-.591	-.619
.350	-.015	-.136	-.222	-.317	-.532	-.065	-.206	-.343	-.321	-.575
.450	-.074	-.168	-.223	-.222	-.470	-.124	-.244	-.301	-.166	-.516
.550	-.078	-.147	-.167	-.133	-.394	-.116	-.227	-.274	-.182	-.461
.650	-.074	-.125	-.124	-.115	-.334	-.096	-.187	-.214	-.175	-.395
.750	-.031	-.061	-.043	-.069	-.265	-.048	-.103	-.102	-.119	-.319
.850	.030	.021	.031	-.010	-.195	.033	.011	.010	-.033	-.273
.900	.064	.062	.070	.027	-.146	.076	.068	.067	.019	-.232
Right side										
.025	-.324	.024	.331	.602	.833	-.893	-.042	.364	.595	.760
.075	-.393	-.089	.151	.368	.575	-.542	-.125	.162	.367	.552
.150	-.330	-.123	.065	.233	.409	-.373	-.142	.065	.240	.403
.250	-.241	-.115	.022	.159	.304	-.296	-.163	-.007	.138	.284
.350	-.222	-.143	-.035	.072	.194	-.297	-.194	-.064	.059	.190
.450	-.207	-.166	-.087	.000	.093	-.276	-.217	-.113	-.013	.096
.550	-.159	-.156	-.099	-.028	.048	-.255	-.210	-.126	-.045	.041
.650	-.092	-.101	-.063	-.021	.015	-.185	-.171	-.099	-.048	.011
.750	-.027	-.053	-.032	-.016	-.002	-.077	-.084	-.035	-.025	-.007
.850	.052	.033	.038	.035	-.002	.035	.034	.045	.016	-.025
.900	.083	.066	.057	.043	-.021	.076	.062	.059	.022	-.073
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.551	.785	.618	.288	-.123	.616	.833	.534	.117	-.202
.025	.362	-.100	-1.051	-1.350	-.716	.481	-.030	-.970	-1.091	-.410
.075	.188	-.146	-.869	-1.211	-.727	.364	-.045	-.903	-.449	-.366
.150	.080	-.174	-.394	-1.075	-.734	.217	-.109	-.635	-.460	-.387
.250	.014	-.181	-.392	-.620	-.692	.063	-.214	-.515	-.458	-.402
.350	-.044	-.217	-.272	-.357	-.430					
.450	-.101	-.276	-.383	-.281	-.409					
.550	-.140	-.305	-.440	-.294	-.413					
.650	-.130	-.273	-.312	-.299	-.422	-.204	-.409	-.425	-.410	-.383
.750	-.078	-.174	-.171	-.270	-.432	-.162	-.213	-.203	-.394	-.379
.850	-.009	-.021	-.028	-.224	-.425	-.084	-.084	-.075	-.355	-.359
.900	.026	.025	.024	-.194	-.415	-.034	-.032	-.027	-.330	-.346
Right side										
.025	-1.036	-.050	.390	.602	.742	-1.057	.071	.510	.706	.815
.075	-.805	-.135	.178	.384	.554	-.847	-.025	.351	.566	.728
.150	-.346	-.132	.089	.264	.421	-.476	-.077	.228	.440	.619
.250	-.352	-.163	.011	.162	.309					
.350	-.306	-.207	-.050	.080	.221					
.450	-.370	-.252	-.101	.019	.144					
.550	-.349	-.285	-.141	-.039	.068					
.650	-.320	-.268	-.132	-.067	.009	-.379	-.367	-.182	-.133	-.053
.750	-.139	-.158	-.083	-.081	-.049	-.185	-.201	-.151	-.207	-.166
.850	.006	.014	.019	-.043	-.074	-.046	-.050	-.039	-.159	-.162
.900	.058	.057	.049	-.051	-.125	.006	-.002	-.008	-.184	-.214

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$ - Continued

(e) $\alpha = 0°$; $M = 0.92$

$\dfrac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.787	.691	.634	.466	.106	.614	.802	.687	.334	-.109
.025	.283	-.054	-.443	-.845	-.054	.325	-.077	-.904	-1.209	-.603
.075	.156	-.090	-.367	-.722	-.744	.148	-.150	-.670	-1.097	-.610
.150	.047	-.155	-.363	-.639	-.692	.055	-.170	-.412	-1.002	-.630
.250	.018	-.132	-.271	-.530	-.613	-.018	-.185	-.327	-.643	-.626
.350	-.032	-.155	-.231	-.354	-.532	-.084	-.226	-.351	-.498	-.580
.450	-.096	-.192	-.239	-.276	-.469	-.150	-.269	-.356	-.156	-.515
.550	-.105	-.180	-.179	-.139	-.395	-.150	-.253	-.247	-.139	-.461
.650	-.108	-.164	-.130	-.105	-.335	-.134	-.241	-.237	-.160	-.399
.750	-.057	-.086	-.050	-.067	-.273	-.074	-.142	-.112	-.122	-.326
.850	.010	.007	.033	-.012	-.207	.013	-.001	.008	-.041	-.276
.900	.049	.054	.069	.021	-.164	.062	.069	.012		-.244
Right side										
.025	-.302	.020	.337	.603	.831	-.830	-.045	.361	.594	.763
.075	-.390	-.090	.153	.366	.575	-.609	-.134	.161	.365	.550
.150	-.356	-.129	.066	.237	.409	-.409	-.154	.063	.234	.401
.250	-.272	-.129	.021	.159	.304	-.317	-.177	-.009	.132	.279
.350	-.245	-.162	-.039	.070	.192	-.335	-.213	-.070	.049	.186
.450	-.238	-.189	-.093	-.002	.087	-.325	-.243	-.122	-.023	.092
.550	-.183	-.184	-.106	-.039	.043	-.250	-.236	-.142	-.058	.034
.650	-.111	-.126	-.071	-.035	.004	-.230	-.215	-.114	-.060	.004
.750	-.046	-.079	-.045	-.033	-.012	-.103	-.126	-.045	-.038	-.016
.850	.038	.019	.030	.020	-.010	.021	.021	.038	.005	-.035
.900	.073	.053	.052	.032	-.036	.065	.053	.055	.008	-.083
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.577	.786	.640	.321	-.077	.637	.834	.561	.157	-.176
.025	.339	-.112	-1.009	-1.292	-.718	.446	-.045	-.981	-1.060	-.383
.075	.169	-.163	-.850	-1.164	-.729	.345	-.052	-.883	-.462	-.367
.150	.060	-.192	-.556	-1.071	-.731	.195	-.113	-.781	-.487	-.391
.250	-.006	-.197	-.380	-.858	-.702	.039	-.210	-.618	-.501	-.407
.350	-.067	-.230	-.366	-.368	-.428					
.450	-.126	-.291	-.312	-.239	-.407					
.550	-.182	-.353	-.430	-.283	-.417					
.650	-.179	-.337	-.391	-.312	-.430	-.308	-.517	-.516	-.418	-.388
.750	-.123	-.262	-.197	-.287	-.440	-.192	-.298	-.241	-.398	-.387
.850	-.024	-.030	-.029	-.238	-.433	-.092	-.124	-.092	-.358	-.364
.900	.011	.028	.030	-.205	-.429	-.053	-.061	-.034	-.335	-.353
Right side										
.025	-.952	-.052	.384	.598	.744	-.944	.071	.507	.705	.819
.075	-.780	-.145	.175	.378	.552	-.848	-.028	.348	.561	.729
.150	-.477	-.146	.086	.257	.418	-.688	-.082	.225	.434	.625
.250	-.372	-.179	.007	.158	.306					
.350	-.351	-.217	-.056	.074	.216					
.450	-.314	-.269	-.106	.011	.137					
.550	-.419	-.328	-.156	-.052	.062					
.650	-.374	-.326	-.148	-.084	.006	-.502	-.470	-.241	-.160	-.064
.750	-.213	-.262	-.100	-.102	-.061	-.254	-.300	-.158	-.237	-.184
.850	-.003	.006	.015	-.060	-.080	-.079	-.108	-.037	-.175	-.165
.900	.052	.058	.045	-.072	-.135	-.020	-.041	-.005	-.202	-.217

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$ - Continued

(f) $\alpha = 9.4°$; M = 0.60

| | C_p for $z/b_v = 0.11$ | | | | | C_p for $z/b_v = 0.38$ | | | | |
$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side									
.000	.442	.821	.473	-.681	-1.137	.132	.691	.105	-.939	-1.220
.025	-.004	-.129	-.258	-.794	-2.065	.355	-.095	-.782	-1.563	-.994
.075	.034	-.118	-.318	-.549	-1.040	.173	-.129	-.486	-1.388	-.966
.150	.000	-.141	-.318	-.455	-.607	.089	-.132	-.357	-.699	-.920
.250	.002	-.093	-.228	-.324	-.420	.025	-.134	-.297	-.340	-.858
.350	-.014	-.102	-.205	-.268	-.347	-.020	-.152	-.267	-.324	-.722
.450	-.032	-.111	-.198	-.238	-.342	-.052	-.159	-.242	-.287	-.575
.550	-.032	-.091	-.159	-.192	-.273	-.036	-.118	-.182	-.234	-.462
.650	-.007	-.057	-.113	-.153	-.211	-.016	-.084	-.134	-.178	-.388
.750	.009	-.025	-.063	-.093	-.158	.009	-.029	-.070	-.107	-.317
.850	.048	.037	.004	-.019	-.091	.055	.034	.018	-.024	-.231
.900	.071	.066	.043	.017	-.045	.082	.068	.057	.020	-.171
	Right side									
.025	-.194	-.036	.022	.534	.815	-.681	-.052	.395	.612	.718
.075	-.328	-.127	.038	.349	.545	-.451	-.107	.195	.402	.547
.150	-.307	-.129	.015	.239	.393	-.344	-.116	.093	.271	.407
.250	-.239	-.100	.004	.186	.299	-.271	-.122	.036	.179	.296
.350	-.212	-.102	-.010	.107	.202	-.250	-.129	-.001	.117	.225
.450	-.198	-.111	-.033	.050	.123	-.225	-.134	-.028	.061	.153
.550	-.150	-.093	-.026	.031	.089	-.180	-.113	-.028	.036	.107
.650	-.100	-.050	-.005	.031	.075	-.123	-.075	-.005	.041	.091
.750	-.052	-.009	.022	.045	.070	-.050	-.013	.027	.057	.075
.850	.023	.048	.064	.073	.080	.032	.057	.068	.087	.068
.900	.055	.068	.073	.082	.082	.059	.073	.080	.080	.038

| | C_p for $z/b_v = 0.66$ | | | | | C_p for $z/b_v = 0.93$ | | | | |
$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side									
.000	.025	.650	-.051	-.635	-.851	.169	.666	-.198	-.621	-.443
.025	.378	-.122	-1.044	-.992	-.632	.506	-.032	-.920	-.773	-.443
.075	.207	-.127	-.578	-.948	-.628	.424	.032	-.684	-.549	-.453
.150	.112	-.132	-.408	-.895	-.616	.337	.021	-.440	-.542	-.480
.250	.062	-.120	-.316	-.780	-.582	.246	-.020	-.272	-.517	-.483
.350	.030	-.125	-.279	-.616	-.561					
.450	.009	-.129	-.251	-.459	-.538					
.550	-.016	-.125	-.215	-.351	-.506	-.004	-.109	-.187	-.289	-.407
.650	.005	-.079	-.157	-.257	-.478	-.014	-.079	-.129	-.243	-.386
.750	.025	-.034	-.084	-.181	-.450	.000	-.032	-.065	-.197	-.354
.850	.046	.014	-.015	-.114	-.402	.021	.003	-.015	-.171	-.328
.900	.064	.041	.018	-.077	-.388					
	Right side									
.025	-.922	-.036	.425	.605	.670	-.925	.093	.556	.653	.690
.075	-.503	-.102	.222	.418	.541	-.581	.071	.441	.614	.732
.150	-.348	-.100	.133	.303	.428	-.314	.071	.374	.568	.723
.250	-.296	-.109	.075	.223	.345					
.350	-.269	-.111	.041	.165	.269					
.450	-.230	-.109	.018	.128	.216					
.550	-.189	-.100	.015	.091	.163	-.164	-.072	.027	.087	.119
.650	-.134	-.063	.025	.080	.119	-.116	-.066	-.005	.011	-.010
.750	-.071	-.009	.036	.066	.068	-.023	.005	.036	-.003	-.065
.850	.018	.046	.080	.077	.017	.014	.030	.048	-.031	-.112
.900	.052	.068	.091	.075	-.036					

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TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$ - Continued

(i) $\alpha = 9.7°$; M = 0.90

C_p for - ($z/b_v = 0.11$)

Left side

$\frac{x}{C_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.344	.925	.460	-.160	
.025	-.081	-.088	-.039	-.455	
.075	.036	-.106	-.257	-.488	
.150	.014	-.159	-.410	-.577	
.250	.001	-.117	-.301	-.481	
.350	-.030	-.127	-.262	-.338	
.450	-.057	-.162	-.274	-.293	
.550	-.048	-.127	-.199	-.243	
.650	-.042	-.100	-.155	-.203	
.750	-.013	-.055	-.090	-.158	
.850	.040	.024	-.004	-.073	
.900	.077	.067	.047	-.019	

Right side

$\frac{x}{C_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	.045	.021	-.025	.336	
.075	-.262	-.107	.053	.315	
.150	-.376	-.127	.046	.252	
.250	-.274	-.103	.014	.187	
.350	-.262	-.117	-.015	.106	
.450	-.269	-.151	-.047	.027	
.550	-.201	-.126	-.046	.004	
.650	-.129	-.091	-.031	-.009	
.750	-.068	-.043	.004	.012	
.850	.021	.039	.062	.047	
.900	.060	.066	.077	.045	

C_p for - ($z/b_v = 0.38$)

Left side

$\frac{x}{C_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.536	.764	.575	.147	
.025	.335	-.092	-1.001	-1.342	
.075	.163	-.141	-.667	-1.196	
.150	.084	-.143	-.463	-.783	
.250	.018	-.159	-.332	-.519	
.350	-.042	-.198	-.348	-.424	
.450	-.088	-.224	-.323	-.396	
.550	-.072	-.183	-.250	-.301	
.650	-.052	-.142	-.178	-.246	
.750	-.013	-.067	-.083	-.172	
.850	.052	.029	.020	-.081	
.900	.092	.078	.075	-.026	

Right side

$\frac{x}{C_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	-.908	-.023	.397	.608	
.075	-.563	-.099	.197	.390	
.150	-.395	-.117	.105	.268	
.250	-.309	-.135	.038	.174	
.350	-.325	-.162	-.012	.105	
.450	-.288	-.177	-.051	.041	
.550	-.244	-.161	-.052	.014	
.650	-.164	-.119	-.028	.012	
.750	-.060	-.034	.014	.024	
.850	.040	.052	.074	.044	
.900	.080	.070	.080	.035	

$z/b_v = 0.66$

Left side

$\frac{x}{C_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.433	.711	.450	.027	
.025	.356	-.138	-1.143	-.875	
.075	.195	-.151	-.970	-.830	
.150	.101	-.165	-.464	-.797	
.250	.049	-.158	-.387	-.702	
.350	.013	-.173	-.295	-.554	
.450	-.021	-.202	-.348	-.447	
.550	-.057	-.212	-.312	-.374	
.650	-.045	-.165	-.203	-.320	
.750	-.010	-.082	-.104	-.269	
.850	.032	.000	-.009	-.220	
.900	.060	.041	.039	-.187	

Right side

$\frac{x}{C_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	-1.092	-.020	.416	.608	
.075	-.868	-.108	.219	.410	
.150	-.348	-.106	.140	.305	
.250	-.374	-.127	.077	.222	
.350	-.284	-.143	.042	.164	
.450	-.319	-.165	.008	.114	
.550	-.283	-.171	-.016	.067	
.650	-.209	-.138	-.015	.040	
.750	-.099	-.063	.008	.014	
.850	.022	.043	.070	.013	
.900	.071	.066	.085	-.013	

$z/b_v = 0.93$

Left side

$\frac{x}{C_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.453	.743	.272	-.265	
.025	.511	-.016	-1.023	-1.095	
.075	.446	.012	-.892	-.461	
.150	.340	-.017	-.759	-.450	
.250	.211	-.092	-.588	-.459	
.650	-.081	-.208	-.233	-.404	
.750	-.062	-.129	-.141	-.370	
.850	-.018	-.043	-.056	-.308	
.900	.009	.001	-.009	-.270	

Right side

$\frac{x}{C_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	-1.171	.134	.573	.731	
.075	-.774	.083	.473	.666	
.150	-.508	.062	.390	.591	
.650	-.234	-.163	-.035	.010	
.750	-.139	-.119	-.051	-.081	
.850	-.023	-.004	.026	-.071	
.900	.022	.028	.040	-.107	

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$ - Continued

(j) $\alpha = 9.7°$; $M = 0.92$

$\dfrac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.344	.945	.474	-.130		.565	.757	.596	.180	
.025	-.067	-.047	.024	-.390		.339	-.049	-.919	-1.329	
.075	.051	-.064	-.183	-.437		.175	-.103	-.466	-1.194	
.150	.035	-.123	-.344	-.543		.086	-.113	-.404	-.777	
.250	.018	-.095	-.269	-.503		.023	-.140	-.318	-.507	
.350	-.020	-.111	-.249	-.377		-.041	-.186	-.348	-.366	
.450	-.057	-.157	-.278	-.289		-.095	-.222	-.352	-.397	
.550	-.054	-.132	-.211	-.240		-.080	-.187	-.254	-.310	
.650	-.049	-.112	-.159	-.208		-.066	-.148	-.193	-.256	
.750	-.025	-.066	-.091	-.161		-.027	-.076	-.093	-.189	
.850	.029	.018	-.012	-.085		.041	.023	.013	-.097	
.900	.065	.056	.037	-.034		.080	.072	.069	-.044	
Right side										
.025	.123	.064	.000	.257		-.824	-.002	.408	.617	
.075	-.185	-.064	.072	.312		-.468	-.071	.208	.400	
.150	-.318	-.092	.069	.272		-.355	-.088	.115	.272	
.250	-.246	-.078	.042	.203		-.291	-.113	.045	.176	
.350	-.247	-.103	.003	.113		-.324	-.152	-.007	.100	
.450	-.271	-.145	-.040	.031		-.303	-.176	-.052	.037	
.550	-.206	-.125	-.048	-.001		-.252	-.161	-.058	.004	
.650	-.140	-.098	-.040	-.015		-.178	-.124	-.034	.001	
.750	-.075	-.055	-.005	.004		-.075	-.057	.004	.009	
.850	.012	.035	.050	.032		.029	.051	.065	.031	
.900	.048	.060	.062	.032		.069	.068	.072	.015	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.450	.709	.460	.048		.479	.738	.307	-.225	
.025	.352	-.103	-1.062	-.932		.498	.020	-.903	-1.065	
.075	.193	-.121	-.901	-.851		.430	.031	-.841	-.455	
.150	.097	-.141	-.396	-.820		.326	-.004	-.706	-.446	
.250	.044	-.143	-.383	-.716		.195	-.080	-.559	-.450	
.350	.006	-.165	-.351	-.552						
.450	-.033	-.197	-.332	-.439						
.550	-.070	-.210	-.358	-.375						
.650	-.058	-.176	-.215	-.333		-.107	-.223	-.241	-.418	
.750	-.026	-.092	-.111	-.289		-.080	-.135	-.151	-.387	
.850	.019	-.006	-.016	-.244		-.030	-.047	-.052	-.339	
.900	.047	.039	.032	-.218		.003	-.002	-.009	-.309	
Right side										
.025	-1.026	-.008	.416	.601		-1.113	.135	.568	.730	
.075	-.726	-.094	.221	.408		-.730	.088	.466	.659	
.150	-.348	-.092	.142	.301		-.451	.060	.381	.585	
.250	-.367	-.116	.077	.221						
.350	-.293	-.135	.037	.163						
.450	-.318	-.164	.001	.109						
.550	-.297	-.177	-.024	.059						
.650	-.235	-.148	-.028	.028		-.254	-.178	-.049	-.007	
.750	-.112	-.092	-.007	-.009		-.150	-.125	-.066	-.105	
.850	.015	.048	.061	-.007		-.031	-.010	.016	-.093	
.900	.063	.066	.076	-.034		.015	.023	.032	-.134	

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$ - Continued

(k) $\alpha = 15.6°$; $M = 0.60$

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	-.373	.829	.223	.093	-.424	.299	.730	.282	-.475	-.932
.025	-.606	-.116	.315	.114	-.433	.124	-.100	-.328	-.698	-.951
.075	-.158	-.129	.029	-.083	-.389	.054	-.141	-.303	-.517	-.816
.150	-.040	-.155	-.130	-.191	-.382	.050	-.136	-.264	-.402	-.663
.250	.017	-.118	-.139	-.179	-.324	.040	-.141	-.250	-.345	-.545
.350	.033	-.095	-.160	-.189	-.287	.020	-.155	-.241	-.303	-.449
.450	.038	-.113	-.171	-.186	-.252	.010	-.166	-.220	-.262	-.377
.550	.038	-.097	-.139	-.150	-.201	.022	-.132	-.178	-.216	-.317
.650	.045	-.067	-.095	-.111	-.152	.029	-.102	-.135	-.170	-.261
.750	.047	-.033	-.059	-.074	-.115	.038	-.049	-.077	-.106	-.192
.850	.057	.018	-.010	-.021	-.062	.057	.015	-.012	-.051	-.129
.900	.061	.045	.013	.006	-.025	.070	.041	.022	-.012	-.087
	Right side									
.025	.341	-.113	-.759	-.271	-.097	-.385	-.088	.112	.169	.168
.075	.001	-.175	-.234	-.072	.029	-.311	-.127	.052	.121	.223
.150	-.131	-.150	-.054	.006	.100	-.274	-.129	.043	.114	.254
.250	-.151	-.113	.020	.059	.156	-.246	-.127	.052	.119	.272
.350	-.167	-.095	.038	.080	.182	-.232	-.136	.045	.116	.268
.450	-.179	-.118	.038	.089	.179	-.214	-.141	.029	.110	.251
.550	-.142	-.100	.043	.087	.154	-.172	-.127	.029	.107	.223
.650	-.094	-.061	.045	.082	.128	-.131	-.086	.043	.103	.191
.750	-.057	-.024	.050	.084	.110	-.068	-.040	.052	.098	.161
.850	.001	.025	.066	.084	.093	-.003	.029	.073	.098	.133
.900	.022	.045	.066	.082	.080	.024	.038	.068	.089	.112
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	-.047	.634	.008	-.631	-.781	-.015	.622	-.402	-.838	-.507
.025	.421	-.113	-1.063	-1.269	-.781	.555	.004	-.934	-.686	-.491
.075	.253	-.127	-.582	-1.147	-.753	.498	.066	-.810	-.606	-.454
.150	.163	-.127	-.407	-.838	-.714	.438	.064	-.508	-.574	-.463
.250	.121	-.116	-.310	-.507	-.654	.338	.029	-.314	-.523	-.449
.350	.089	-.120	-.273	-.368	-.577					
.450	.068	-.123	-.245	-.285	-.496					
.550	.050	-.118	-.204	-.239	-.415					
.650	.047	-.081	-.144	-.166	-.331	.045	-.109	-.174	-.205	-.264
.750	.061	-.035	-.079	-.115	-.280	.024	-.081	-.123	-.157	-.217
.850	.063	.004	-.017	-.053	-.220	.029	-.028	-.052	-.106	-.187
.900	.070	.027	.015	-.028	-.194	.043	-.005	-.015	-.083	-.166
	Right side									
.025	-.990	-.049	.451	.591	.571	-1.068	.082	.568	.649	.655
.075	-.562	-.109	.259	.403	.476	-.669	.110	.499	.635	.722
.150	-.366	-.095	.176	.300	.407	-.357	.121	.469	.621	.718
.250	-.299	-.102	.130	.243	.360					
.350	-.267	-.107	.100	.197	.326					
.450	-.228	-.102	.077	.174	.295					
.550	-.191	-.090	.070	.146	.265					
.650	-.133	-.063	.064	.132	.216	-.131	-.054	.084	.142	.212
.750	-.066	-.031	.066	.116	.172	-.100	-.063	.036	.061	.098
.850	.010	.029	.084	.110	.126	-.017	.004	.059	.052	.056
.900	.043	.052	.093	.098	.087	.010	.027	.059	.038	.008

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0°$ - Continued

(l) $\alpha = 15.8°$; M = 0.80

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.005	.876	.477	.096	-.327	.319	.785	.282	-.285	-.579
.025	-.920	.122	.336	.017	-.444	-.012	-.060	-.326	-.554	-.654
.075	-.513	-.079	.022	-.136	-.400	-.070	-.148	-.326	-.468	-.601
.150	-.241	-.183	-.185	-.241	-.384	-.053	-.152	-.302	-.407	-.525
.250	-.119	-.141	-.178	-.212	-.333	-.030	-.155	-.282	-.366	-.472
.350	-.050	-.100	-.187	-.203	-.283	-.026	-.169	-.276	-.336	-.420
.450	-.009	-.120	-.204	-.195	-.238	-.018	-.178	-.248	-.288	-.357
.550	.016	-.088	-.145	-.140	-.184	.008	-.138	-.202	-.238	-.303
.650	.034	-.047	-.095	-.093	-.141	.020	-.099	-.149	-.185	-.257
.750	.046	-.016	-.058	-.067	-.105	.040	-.047	-.087	-.121	-.195
.850	.060	.030	-.012	-.017	-.049	.063	.021	-.018	-.055	-.137
.900	.068	.060	.014	.009	-.010	.077	.060	.023	-.015	-.081
Right side										
.025	.374	-.765	-.952	-.419	-.294	-.408	-.232	-.043	-.014	.026
.075	-.022	-.449	-.596	-.238	-.171	-.355	-.242	-.084	-.040	.094
.150	-.189	-.238	-.277	-.157	-.050	-.318	-.193	-.064	-.015	.148
.250	-.189	-.134	-.124	-.075	.053	-.284	-.144	-.026	.021	.214
.350	-.200	-.063	-.055	-.014	.117	-.266	-.125	-.009	.045	.233
.450	-.209	-.062	-.012	.029	.154	-.240	-.125	-.001	.064	.242
.550	-.151	-.042	.017	.052	.154	-.197	-.097	.014	.079	.232
.650	-.090	-.015	.034	.067	.139	-.137	-.056	.034	.091	.212
.750	-.048	.015	.048	.070	.148	-.067	-.007	.052	.102	.185
.850	-.001	.056	.068	.087	.106	-.003	.051	.075	.102	.156
.900	.017	.067	.068	.084	.094	.027	.068	.075	.097	.132
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.339	.680	.425	.023	-.375	.278	.684	.057	-.468	-.439
.025	.350	-.088	-1.104	-1.150	-.640	.556	.022	-.871	-.772	-.445
.075	.189	-.126	-.580	-1.006	-.618	.498	.080	-.739	-.601	-.391
.150	.114	-.135	-.422	-.784	-.588	.419	.056	-.615	-.569	-.390
.250	.089	-.125	-.320	-.520	-.545	.305	.008	-.342	-.544	-.381
.350	.074	-.126	-.282	-.383	-.503					
.450	.051	-.137	-.259	-.304	-.442					
.550	.034	-.131	-.216	-.246	-.381					
.650	.039	-.091	-.149	-.169	-.309	.014	-.129	-.188	-.217	-.238
.750	.056	-.044	-.080	-.113	-.266	-.001	-.094	-.124	-.159	-.208
.850	.066	.010	-.003	-.049	-.220	.016	-.028	-.047	-.102	-.168
.900	.077	.042	.028	-.020	-.192	.034	.011	-.006	-.076	-.145
Right side										
.025	-1.185	-.091	.374	.562	.641	-.897	.100	.565	.696	.774
.075	-.562	-.129	.181	.358	.519	-.673	.128	.496	.666	.808
.150	-.387	-.091	.123	.268	.451	-.568	.128	.445	.617	.779
.250	-.324	-.086	.095	.218	.400					
.350	-.287	-.089	.072	.183	.367					
.450	-.249	-.085	.060	.161	.340					
.550	-.206	-.076	.046	.138	.303					
.650	-.145	-.050	.049	.123	.260	-.142	-.053	.048	.123	.248
.750	-.071	-.008	.059	.113	.211	-.105	-.060	.011	.052	.127
.850	.020	.056	.092	.119	.159	-.022	.019	.049	.052	.102
.900	.057	.077	.097	.102	.112	.017	.042	.059	.029	.029

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION; $i_t = 0^0$ - Concluded

(m) $\alpha = 15.9^0$; $M = 0.85$

$\dfrac{x}{c_v}$	\multicolumn{5}{c}{C_p for – $z/b_v = 0.11$}					\multicolumn{5}{c}{C_p for – $z/b_v = 0.38$}				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.312	.905	.652	.075		.372	.781	.290	-.311	
.025	-1.074	.049	.196	-.117		-.016	-.143	-.198	-.632	
.075	-.562	-.157	-.073	-.246		-.126	-.226	-.456	-.543	
.150	-.315	-.276	-.297	-.345		-.133	-.220	-.416	-.484	
.250	-.209	-.219	-.271	-.287		-.116	-.204	-.368	-.427	
.350	-.145	-.159	-.247	-.260		-.110	-.217	-.338	-.380	
.450	-.094	-.169	-.256	-.237		-.093	-.230	-.310	-.337	
.550	-.047	-.119	-.181	-.172		-.050	-.177	-.253	-.274	
.650	-.013	-.075	-.130	-.123		-.024	-.133	-.194	-.219	
.750	.013	-.035	-.087	-.098		.003	-.075	-.130	-.150	
.850	.037	.021	-.034	-.038		.040	.002	-.053	-.078	
.900	.050	.046	-.007	-.011		.057	.049	-.013	-.033	
Right side										
.025	.282	-.694	-1.007	-.489		-.618	-.280	-.081	-.096	
.075	-.100	-.882	-.668	-.330		-.504	-.308	-.168	-.126	
.150	-.312	-.323	-.341	-.246		-.431	-.254	-.163	-.106	
.250	-.279	-.229	-.238	-.169		-.352	-.199	-.131	-.059	
.350	-.258	-.149	-.167	-.102		-.318	-.179	-.101	-.033	
.450	-.250	-.129	-.110	-.046		-.286	-.173	-.078	-.004	
.550	-.179	-.085	-.058	-.006		-.233	-.142	-.051	.024	
.650	-.112	-.043	-.021	.022		-.172	-.090	-.018	.049	
.750	-.066	-.012	.003	.014		-.093	-.035	.014	.071	
.850	-.007	.041	.042	.063		-.020	.035	.049	.086	
.900	.009	.052	.047	.062		.013	.051	.049	.075	

$\dfrac{x}{c_v}$	\multicolumn{5}{c}{$z/b_v = 0.66$}					\multicolumn{5}{c}{$z/b_v = 0.93$}				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.378	.669	.460	.160		.315	.689	.154	-.350	
.025	.299	-.139	-1.228	-1.174		.528	.005	-1.140	-.833	
.075	.134	-.176	-.737	-1.009		.475	.056	-.766	-.615	
.150	.059	-.183	-.497	-.794		.386	.022	-.633	-.566	
.250	.034	-.166	-.363	-.573		.257	-.035	-.436	-.558	
.350	.020	-.166	-.328	-.445						
.450	-.003	-.177	-.308	-.360						
.550	-.019	-.170	-.263	-.290						
.650	-.009	-.132	-.186	-.214		-.037	-.179	-.218	-.264	
.750	.014	-.080	-.107	-.147		-.046	-.137	-.153	-.202	
.850	.037	-.012	-.021	-.090		-.016	-.058	-.071	-.145	
.900	.051	.025	.013	-.051		.007	-.012	-.027	-.120	
Right side										
.025	-1.235	-.132	.309	.529		-1.104	.072	.539	.697	
.075	-.761	-.183	.112	.313		-.705	.096	.469	.659	
.150	-.422	-.146	.057	.226		-.614	.093	.410	.605	
.250	-.352	-.133	.036	.180						
.350	-.331	-.126	.014	.146						
.450	-.285	-.126	.006	.126						
.550	-.243	-.119	-.010	.108						
.650	-.173	-.092	.002	.096		-.172	-.105	.009	.092	
.750	-.092	-.045	.017	.083		-.132	-.100	-.036	.014	
.850	.009	.038	.062	.091		-.037	-.012	.019	.025	
.900	.049	.063	.069	.078		.003	.026	.026	-.006	

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$

(a) $\alpha = 0°$; $M = 0.60$

$\frac{x}{c_v}$	C_p for—					C_p for—				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.607	.665	.360	-.480	-1.371	.306	.748	.335	-.433	-1.234
.025	.272	-.098	-.532	-1.152	-2.378	.332	-.091	-.624	-1.087	-1.186
.075	.131	-.109	-.371	-.695	-1.544	.144	-.128	-.428	-.960	-1.154
.150	.038	-.137	-.297	-.487	-.671	.052	-.139	-.318	-.714	-1.105
.250	.022	-.095	-.201	-.308	-.433	-.013	-.146	-.267	-.442	-1.043
.350	-.032	-.112	-.175	-.245	-.348	-.064	-.160	-.249	-.343	-.876
.450	-.064	-.123	-.162	-.208	-.301	-.103	-.165	-.231	-.285	-.659
.550	-.055	-.102	-.127	-.169	-.318	-.099	-.144	-.175	-.222	-.498
.650	-.038	-.077	-.086	-.127	-.248	-.078	-.107	-.120	-.159	-.375
.750	-.022	-.036	-.040	-.092	-.170	-.038	-.047	-.061	-.094	-.262
.850	.019	.011	.008	-.034	-.082	.019	.020	.022	-.009	-.144
.900	.040	.045	.045	.010	-.034	.050	.057	.052	.033	-.087
	Right side									
.025	-.460	-.029	.328	.606	.830	-.631	-.063	.379	.604	.740
.075	-.381	-.109	.130	.353	.551	-.427	-.118	.160	.363	.535
.150	-.302	-.125	.052	.212	.378	-.325	-.132	.064	.223	.375
.250	-.207	-.100	.020	.138	.271	-.261	-.139	-.003	.121	.244
.350	-.180	-.107	-.019	.070	.172	-.233	-.146	-.049	.052	.156
.450	-.161	-.125	-.061	.017	.105	-.182	-.135	-.079	-.025	.043
.550	-.119	-.093	-.040	.019	.087	-.175	-.132	-.079	-.018	.045
.650	-.080	-.059	-.021	.026	.077	-.113	-.091	-.047	-.004	.036
.750	-.034	-.022	.004	.045	.077	-.050	-.033	-.010	.024	.036
.850	.029	.031	.050	.075	.091	.033	.043	.054	.066	.054
.900	.054	.059	.059	.073	.087	.063	.061	.061	.061	.043
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.216	.720	.264	-.429	-.701	.408	.681	.231	-.329	-.505
.025	.350	-.123	-.824	-1.092	-.759	.214	-.211	-.736	-.851	-.537
.075	.154	-.144	-.486	-.990	-.740	.031	-.162	-.382	-.837	-.549
.150	.040	-.158	-.364	-.804	-.726	-.057	-.151	-.284	-.726	-.604
.250	-.032	-.165	-.286	-.552	-.694	-.103	-.162	-.251	-.540	-.599
.350	-.085	-.174	-.254	-.377	-.659					
.450	-.117	-.176	-.226	-.292	-.625					
.550	-.108	-.151	-.187	-.229	-.585					
.650	-.085	-.116	-.125	-.166	-.528	-.103	-.091	-.120	-.347	-.410
.750	-.050	-.047	-.056	-.094	-.465	-.071	-.038	-.079	-.329	-.382
.850	.008	.008	.002	-.032	-.387	-.029	-.003	-.061	-.299	-.357
.900	.029	.029	.027	-.004	-.359	-.008	.017	-.049	-.264	-.341
	Right side									
.025	-.726	-.072	.392	.597	.701	-.640	-.135	.268	.423	.498
.075	-.483	-.135	.151	.358	.491	-.367	-.165	.031	.142	.246
.150	-.339	-.137	.066	.216	.343	-.279	-.148	-.047	.005	.096
.250	-.284	-.155	-.012	.098	.211					
.350	-.258	-.169	-.077	.022	.112					
.450	-.221	-.162	-.100	-.032	.040					
.550	-.175	-.142	-.102	-.053	.003					
.650	-.117	-.100	-.067	-.041	-.015	.031	-.001	-.042	-.055	-.080
.750	-.052	-.042	-.031	-.006	-.027	-.080	-.033	-.054	-.074	-.121
.850	.029	.036	.041	.047	-.024	-.022	.015	-.003	-.032	-.124
.900	.056	.054	.061	.052	-.057	-.008	.031	.011	-.027	-.147

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(b) $\alpha = 0°$; M = 0.80

$\frac{x}{c_v}$	C_p for — $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for — $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.717	.677	.545	.183	-.324	.490	.781	.545	.045	-.586
.025	.281	-.055	-.527	-1.170	-1.565	.335	-.070	-.955	-1.686	-1.006
.075	.146	-.088	-.371	-.909	-1.284	.152	-.117	-.449	-1.420	-1.022
.150	.043	-.123	-.324	-.529	-1.007	.055	-.132	-.344	-.597	-.961
.250	.017	-.090	-.214	-.306	-.623	-.012	-.147	-.278	-.400	-.877
.350	-.024	-.109	-.187	-.251	-.399	-.075	-.168	-.268	-.344	-.774
.450	-.072	-.126	-.173	-.223	-.329	-.121	-.183	-.246	-.292	-.656
.550	-.063	-.102	-.133	-.175	-.304	-.109	-.151	-.185	-.219	-.543
.650	-.047	-.072	-.092	-.141	-.232	-.084	-.108	-.123	-.149	-.445
.750	-.027	-.037	-.043	-.098	-.188	-.035	-.036	-.054	-.069	-.352
.850	.016	.026	.022	-.028	-.119	.028	.044	.035	.019	-.249
.900	.042	.068	.058	.016	-.067	.068	.088	.071	.059	-.189
	Right side									
.025	-.415	.017	.338	.601	.827	-.958	-.031	.377	.604	.751
.075	-.392	-.084	.140	.354	.566	-.429	-.100	.163	.364	.541
.150	-.310	-.109	.059	.221	.389	-.331	-.117	.065	.227	.388
.250	-.210	-.085	.024	.149	.287	-.268	-.133	-.004	.128	.262
.350	-.187	-.108	-.025	.071	.184	-.245	-.153	-.051	.050	.174
.450	-.167	-.120	-.065	.015	.105	-.190	-.138	-.089	-.020	.056
.550	-.118	-.085	-.048	.015	.086	-.178	-.138	-.098	-.031	.034
.650	-.070	-.046	-.022	.027	.074	-.106	-.088	-.063	-.014	.022
.750	-.024	-.010	-.002	.036	.063	-.026	-.008	-.011	.018	.017
.850	.037	.050	.042	.065	.074	.059	.068	.059	.068	.027
.900	.068	.077	.058	.073	.060	.083	.089	.077	.077	.008

$\frac{x}{c_v}$	$z/b_v = 0.66$ $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$z/b_v = 0.93$ $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side									
.000	.427	.751	.477	.004	-.431	.502	.710	.390	.015	-.333
.025	.345	-.108	-1.091	-1.194	-.702	.209	-.228	-1.010	-.707	-.476
.075	.161	-.142	-.567	-1.120	-.702	.008	-.183	-.581	-.701	-.490
.150	.040	-.163	-.399	-.959	-.699	-.076	-.159	-.294	-.701	-.542
.250	-.040	-.171	-.312	-.778	-.669	-.124	-.168	-.248	-.617	-.546
.350	-.099	-.184	-.271	-.581	-.630					
.450	-.133	-.189	-.239	-.399	-.588					
.550	-.132	-.162	-.193	-.262	-.543					
.650	-.095	-.109	-.121	-.156	-.499	-.093	-.067	-.118	-.358	-.388
.750	-.038	-.019	-.045	-.072	-.448	-.053	-.007	-.069	-.310	-.369
.850	.019	.038	.024	-.008	-.398	-.003	.038	-.046	-.274	-.339
.900	.042	.067	.051	.021	-.373	.011	.061	-.030	-.237	-.324
	Right side									
.025	-1.060	-.040	.389	.598	.719	-.969	-.135	.254	.419	.530
.075	-.503	-.120	.164	.360	.518	-.461	-.180	-.004	.128	.267
.150	-.353	-.123	.064	.222	.365	-.271	-.154	-.080	-.008	.097
.250	-.302	-.151	-.033	.102	.226					
.350	-.268	-.178	-.091	.010	.115					
.450	-.227	-.171	-.124	-.045	.037					
.550	-.178	-.141	-.121	-.065	-.016					
.650	-.106	-.088	-.080	-.048	-.036	.013	.007	-.040	-.055	-.097
.750	-.033	-.020	-.033	-.017	-.053	-.055	-.008	-.051	-.068	-.133
.850	.054	.070	.058	.050	-.041	.005	.061	.016	-.020	-.122
.900	.091	.100	.083	.064	-.079	.016	.076	.029	-.023	-.145

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(c) $\alpha = 0°$; M = 0.85

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.742	.680	.587	.301	-.144	.537	.787	.600	.187	-.403
.025	.280	-.059	-.528	-1.095	-1.341	.326	-.079	-1.088	-1.559	-.914
.075	.142	-.095	-.384	-.874	-1.153	.144	-.136	-.426	-1.395	-.937
.150	.043	-.145	-.351	-.670	-.973	.050	-.155	-.356	-1.209	-.930
.250	.017	-.109	-.228	-.336	-.716	-.026	-.169	-.304	-.336	-.858
.350	-.036	-.131	-.198	-.255	-.516	-.090	-.194	-.290	-.293	-.757
.450	-.090	-.152	-.189	-.211	-.407	-.146	-.209	-.267	-.271	-.648
.550	-.074	-.122	-.142	-.159	-.338	-.131	-.179	-.192	-.211	-.556
.650	-.063	-.089	-.101	-.134	-.256	-.106	-.132	-.131	-.142	-.470
.750	-.040	-.051	-.046	-.088	-.216	-.046	-.048	-.053	-.064	-.380
.850	.007	.012	.018	-.015	-.134	.030	.032	.034	.029	-.276
.900	.039	.055	.058	.030	-.076	.067	.076	.076	.069	-.217
	Right side									
.025	-.388	.012	.338	.608	.833	-.986	-.049	.371	.598	.750
.075	-.397	-.097	.141	.359	.563	-.436	-.124	.159	.360	.543
.150	-.328	-.129	.054	.223	.399	-.354	-.139	.058	.227	.386
.250	-.226	-.108	.015	.155	.290	-.288	-.156	.012	.125	.272
.350	-.200	-.128	-.040	.072	.187	-.268	-.175	-.069	.046	.169
.450	-.184	-.142	-.078	.006	.103	-.204	-.165	-.103	-.025	.056
.550	-.127	-.106	-.066	.003	.079	-.184	-.166	-.116	-.042	.029
.650	-.080	-.064	-.038	.025	.072	-.116	-.108	-.079	-.029	.007
.750	-.027	-.029	-.019	.033	.054	-.030	-.027	-.019	.009	.009
.850	.037	.039	.038	.068	.064	.056	.062	.054	.069	.014
.900	.070	.062	.056	.068	.050	.089	.079	.078	.079	-.016
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.470	.757	.533	.125	-.331	.522	.711	.435	.010	-.306
.025	.336	-.114	-1.210	-1.641	-.746	.187	-.262	-1.089	-.975	-.504
.075	.154	-.158	-.579	-1.473	-.758	-.024	-.218	-.678	-.907	-.530
.150	.033	-.188	-.420	-1.291	-.758	-.106	-.188	-.310	-.865	-.593
.250	-.057	-.198	-.321	-.500	-.700	-.141	-.201	-.261	-.706	-.581
.350	-.121	-.216	-.278	-.355	-.666					
.450	-.161	-.221	-.254	-.285	-.620					
.550	-.154	-.186	-.199	-.199	-.570					
.650	-.108	-.132	-.128	-.122	-.514	-.108	-.082	-.128	-.396	-.414
.750	-.043	-.028	-.045	-.051	-.450	-.050	-.022	-.078	-.350	-.393
.850	.020	.028	.030	.020	-.387	-.006	.029	-.053	-.301	-.366
.900	.047	.060	.058	.042	-.366	.013	.053	-.032	-.262	-.350
	Right side									
.025	-1.163	-.058	.382	.588	.717	-1.138	-.164	.245	.423	.530
.075	-.436	-.144	.154	.360	.516	-.490	-.218	-.032	.119	.267
.150	-.371	-.148	.057	.217	.364	-.270	-.189	-.106	-.041	.086
.250	-.314	-.179	-.043	.095	.226					
.350	-.281	-.213	-.108	.000	.107					
.450	-.241	-.204	-.151	-.062	.023					
.550	-.186	-.174	-.138	-.084	-.034					
.650	-.111	-.105	-.093	-.055	-.051	-.038	-.049	-.055	-.075	-.109
.750	-.034	-.032	-.042	-.025	-.064	-.060	-.019	-.049	-.068	-.153
.850	.059	.059	.060	.055	-.053	.003	.050	.018	-.018	-.133
.900	.096	.092	.081	.072	-.084	.020	.072	.034	-.018	-.158

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TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(d) $\alpha = 0°$; M = 0.90

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v=0.11$					$z/b_v=0.38$				
					Left side					
.000	.773	.689	.639	.409	.008	.595	.801	.661	.321	-.228
.025	.291	-.048	-.468	-.993	-1.187	.328	-.075	-.993	-1.337	-.802
.075	.154	-.089	-.377	-.782	-1.027	.154	-.137	-.556	-1.213	-.815
.150	.045	-.141	-.355	-.660	-.882	.052	-.157	-.359	-1.104	-.833
.250	.024	-.109	-.227	-.489	-.691	-.022	-.172	-.307	-.620	-.799
.350	-.034	-.136	-.206	-.327	-.557	-.096	-.210	-.325	-.431	-.722
.450	-.099	-.163	-.196	-.262	-.474	-.160	-.231	-.264	-.268	-.636
.550	-.084	-.128	-.136	-.165	-.390	-.154	-.195	-.188	-.154	-.549
.650	-.073	-.097	-.089	-.124	-.317	-.124	-.141	-.119	-.106	-.479
.750	-.049	-.052	-.033	-.078	-.272	-.058	-.043	-.039	-.042	-.408
.850	.005	.019	.039	-.006	-.178	.032	.043	.050	.041	-.324
.900	.038	.058	.071	.034	-.114	.071	.086	.092	.083	-.265
					Right side					
.025	-.339	.030	.348	.605	.837	-.935	-.036	.368	.588	.756
.075	-.411	-.085	.158	.362	.572	-.606	-.117	.166	.357	.542
.150	-.356	-.120	.062	.228	.407	-.375	-.140	.062	.226	.390
.250	-.240	-.105	.024	.151	.296	-.304	-.163	-.010	.123	.268
.350	-.214	-.129	-.036	.067	.190	-.296	-.194	-.071	.035	.171
.450	-.197	-.155	-.085	-.005	.099	-.215	-.182	-.117	-.044	.060
.550	-.125	-.112	-.068	-.005	.079	-.186	-.172	-.128	-.064	.017
.650	-.073	-.070	-.036	.013	.070	-.109	-.113	-.089	-.048	-.001
.750	-.022	-.031	-.016	.021	.045	-.019	-.021	-.020	-.002	-.008
.850	.048	.043	.039	.059	.043	.069	.065	.060	.059	-.005
.900	.076	.069	.058	.067	.020	.100	.092	.082	.078	-.035
	$z/b_v=0.66$					$z/b_v=0.93$				
					Left side					
.000	.526	.769	.597	.253	-.197	.556	.722	.495	.142	-.230
.025	.335	-.113	-1.116	-1.417	-.750	.186	-.287	-1.079	-1.333	-.521
.075	.150	-.163	-.908	-1.278	-.764	-.042	-.260	-1.047	-1.341	-.536
.150	.032	-.199	-.383	-1.202	-.766	-.170	-.210	-.343	-1.084	-.589
.250	-.065	-.219	-.393	-1.112	-.671	-.182	-.225	-.191	-.710	-.580
.350	-.140	-.249	-.239	-.666	-.640					
.450	-.189	-.246	-.239	-.162	-.613					
.550	-.181	-.206	-.195	-.098	-.580					
.650	-.127	-.137	-.115	-.065	-.538	-.113	-.078	-.117	-.323	-.439
.750	-.060	-.025	-.029	-.017	-.481	-.032	-.011	-.055	-.305	-.414
.850	.033	.034	.042	.038	-.415	.013	.043	-.037	-.277	-.386
.900	.052	.067	.074	.057	-.387	.029	.063	-.021	-.240	-.374
					Right side					
.025	-1.077	-.048	.377	.576	.717	-1.068	-.167	.238	.410	.535
.075	-.857	-.147	.158	.342	.517	-.990	-.262	-.060	.099	.273
.150	-.346	-.152	.054	.209	.369	-.245	-.211	-.172	-.085	.082
.250	-.378	-.195	-.047	.079	.229					
.350	-.244	-.233	-.127	-.019	.112					
.450	-.237	-.227	-.172	-.105	.017					
.550	-.189	-.187	-.158	-.124	-.047					
.650	-.105	-.110	-.103	-.082	-.071	-.060	-.062	-.049	-.077	-.126
.750	-.027	-.025	-.036	-.034	-.084	-.054	-.012	-.043	-.074	-.173
.850	.075	.075	.064	.054	-.062	.010	.063	.036	-.011	-.143
.900	.109	.108	.090	.075	-.094	.021	.080	.047	-.009	-.165

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(e) $\alpha = 0°$; $M = 0.92$

$\dfrac{x}{C_v}$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.778	.692	.656	.446	.066	.621	.799	.677	.365	-.168
.025	.289	-.041	-.449	-.953	-1.170	.322	-.073	-.953	-1.269	-.772
.075	.155	-.085	-.378	-.751	-1.003	.149	-.141	-.683	-1.156	-.781
.150	.047	-.143	-.358	-.639	-.855	.049	-.162	-.411	-1.058	-.802
.250	.020	-.111	-.243	-.542	-.666	-.027	-.183	-.314	-.671	-.775
.350	-.039	-.142	-.218	-.370	-.556	-.105	-.226	-.341	-.508	-.718
.450	-.107	-.180	-.219	-.315	-.498	-.174	-.264	-.331	-.424	-.642
.550	-.093	-.143	-.151	-.203	-.416	-.177	-.219	-.182	-.159	-.570
.650	-.084	-.112	-.097	-.130	-.356	-.140	-.159	-.122	-.081	-.508
.750	-.057	-.061	-.032	-.073	-.311	-.065	-.041	-.036	-.020	-.437
.850	.000	.015	.036	-.001	-.216	.030	.038	.052	.057	-.356
.900	.035	.061	.068	.044	-.150	.075	.087	.096	.096	-.302
	Right side									
.025	-.319	.034	.347	.612	.845	-.873	-.036	.363	.590	.759
.075	-.400	-.085	.154	.368	.576	-.635	-.128	.158	.360	.549
.150	-.369	-.125	.062	.233	.410	-.408	-.146	.056	.227	.395
.250	-.255	-.115	.020	.157	.299	-.306	-.170	-.020	.119	.272
.350	-.229	-.141	-.045	.064	.189	-.322	-.210	-.081	.036	.169
.450	-.209	-.171	-.098	-.010	.098	-.258	-.215	-.137	-.051	.060
.550	-.137	-.129	-.082	-.013	.077	-.177	-.200	-.152	-.079	.015
.650	-.072	-.078	-.051	.006	.064	-.102	-.125	-.103	-.060	-.014
.750	-.023	-.036	-.032	.014	.036	-.017	-.022	-.032	-.010	-.021
.850	.050	.041	.036	.058	.027	.072	.066	.052	.061	-.021
.900	.083	.070	.052	.066	-.007	.104	.091	.081	.078	-.058
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.546	.770	.622	.301	-.138	.576	.721	.514	.198	-.200
.025	.326	-.113	-1.077	-1.345	-.726	.173	-.285	-1.047	-1.303	-.519
.075	.148	-.170	-.892	-1.213	-.731	-.054	-.302	-1.023	-1.309	-.537
.150	.025	-.209	-.516	-1.150	-.739	-.205	-.248	-.705	-1.092	-.582
.250	-.071	-.239	-.394	-1.095	-.689	-.224	-.253	-.161	-.964	-.581
.350	-.161	-.294	-.391	-.985	-.654					
.450	-.226	-.279	-.198	-.398	-.609					
.550	-.203	-.234	-.193	-.056	-.579					
.650	-.140	-.142	-.116	-.020	-.550	-.098	-.081	-.119	-.342	-.451
.750	-.051	-.028	-.034	.014	-.495	-.033	-.011	-.064	-.307	-.432
.850	.037	.037	.041	.058	-.436	.008	.041	-.040	-.273	-.405
.900	.054	.072	.074	.079	-.414	.028	.064	-.024	-.236	-.392
	Right side									
.025	-1.017	-.054	.367	.574	.720	-.999	-.175	.227	.411	.544
.075	-.830	-.154	.149	.346	.515	-.971	-.320	-.072	.105	.274
.150	-.441	-.164	.045	.215	.372	-.597	-.248	-.204	-.085	.081
.250	-.386	-.219	-.061	.080	.229					
.350	-.390	-.285	-.152	-.028	.107					
.450	-.196	-.257	-.204	-.119	.005					
.550	-.180	-.209	-.182	-.147	-.062					
.650	-.099	-.116	-.120	-.099	-.083	-.091	-.077	-.068	-.074	-.149
.750	-.021	-.030	-.043	-.040	-.102	-.059	-.014	-.048	-.062	-.188
.850	.080	.078	.065	.058	-.074	.009	.063	.031	.000	-.158
.900	.110	.104	.091	.084	-.111	.021	.075	.047	.006	-.175

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(f) α = 9.4°;   M = 0.60

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.404	.799	.443	-.682	-1.174	.155	.683	.157	-.727	-1.512
.025	-.027	-.136	-.234	-.727	-2.628	.336	-.108	-.727	-1.225	-1.336
.075	.016	-.133	-.318	-.550	-.863	.150	-.145	-.475	-1.032	-1.292
.150	-.022	-.156	-.313	-.457	-.658	.055	-.154	-.352	-.777	-1.372
.250	-.018	-.113	-.229	-.332	-.429	-.011	-.165	-.291	-.466	-1.192
.350	-.047	-.124	-.209	-.282	-.361	-.067	-.181	-.282	-.347	-.801
.450	-.070	-.142	-.202	-.254	-.361	-.102	-.204	-.259	-.302	-.464
.550	-.079	-.133	-.184	-.222	-.295	-.104	-.165	-.209	-.241	-.333
.650	-.058	-.111	-.150	-.191	-.235	-.083	-.129	-.161	-.191	-.285
.750	-.047	-.079	-.109	-.125	-.178	-.054	-.083	-.093	-.118	-.208
.850	-.013	-.031	-.050	-.072	-.105	.005	-.013	-.018	-.032	-.114
.900	.012	.005	-.007	-.027	-.057	.030	.016	.018	.009	-.066
Right side										
.025	-.181	-.054	.005	.498	.813	-.698	-.072	.375	.580	.708
.075	-.333	-.142	.018	.316	.548	-.464	-.129	.168	.366	.543
.150	-.315	-.142	-.004	.212	.388	-.351	-.142	.071	.232	.388
.250	-.238	-.106	-.022	.155	.285	-.287	-.154	.005	.132	.269
.350	-.224	-.122	-.047	.082	.194	-.269	-.165	-.047	.068	.183
.450	-.215	-.138	-.066	.028	.107	-.244	-.183	-.084	.007	.103
.550	-.174	-.117	-.057	.012	.080	-.208	-.163	-.088	-.020	.055
.650	-.142	-.090	-.041	.012	.071	-.147	-.120	-.066	-.020	.039
.750	-.102	-.065	-.022	.012	.050	-.081	-.061	-.032	.003	.036
.850	-.031	-.009	.021	.048	.073	-.002	.007	.028	.055	.059
.900	.003	.019	.039	.053	.071	.028	.023	.032	.041	.043
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.032	.626	.009	-.795	-1.123	.182	.570	-.043	-.463	-.854
.025	.336	-.142	-1.022	-1.325	-.909	.209	-.215	-.888	-.727	-.612
.075	.157	-.172	-.543	-1.297	-.895	.007	-.201	-.482	-.709	-.601
.150	.041	-.188	-.402	-1.193	-.863	-.092	-.188	-.309	-.672	-.610
.250	-.033	-.185	-.318	-.802	-.827	-.077	-.126	-.222	-.575	-.598
.350	-.088	-.197	-.284	-.382	-.797					
.450	-.115	-.199	-.254	-.254	-.749					
.550	-.131	-.190	-.213	-.220	-.676					
.650	-.099	-.142	-.152	-.168	-.591	-.115	-.099	-.143	-.368	-.480
.750	-.056	-.088	-.088	-.100	-.502	-.079	-.054	-.111	-.329	-.450
.850	-.018	-.024	-.027	-.043	-.397	-.047	-.015	-.082	-.284	-.429
.900	.003	-.006	-.004	-.016	-.356	-.024	.005	-.063	-.259	-.413
Right side										
.025	-.931	-.083	.382	.566	.639	-.748	-.149	.246	.371	.445
.075	-.519	-.147	.164	.350	.498	-.437	-.201	-.004	.109	.230
.150	-.360	-.151	.062	.223	.354	-.276	-.172	-.091	-.016	.073
.250	-.312	-.174	-.018	.107	.235					
.350	-.281	-.188	-.079	.025	.132					
.450	-.247	-.190	-.109	-.025	.059					
.550	-.204	-.172	-.116	-.050	.016					
.650	-.147	-.138	-.084	-.043	-.012	-.117	-.088	-.111	-.127	-.149
.750	-.079	-.056	-.043	-.022	-.021	-.092	-.061	-.086	-.118	-.167
.850	.003	.007	.023	.025	-.018	-.036	.005	-.027	-.077	-.167
.900	.028	.025	.043	.039	-.037	-.022	.016	-.013	-.077	-.190

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(g) $\alpha = 9.6°$; $M = 0.80$

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.355	.853	.427	-.364	-.157	.405	.724	.442	-.136	-.807
.025	-.048	-.110	-.142	-.590	-1.622	.324	-.097	-.955	-1.493	-1.070
.075	.017	-.128	-.313	-.566	-1.094	.148	-.142	-.496	-1.149	-1.047
.150	-.016	-.162	-.352	-.533	-.702	.054	-.151	-.370	-.816	-1.006
.250	-.022	-.107	-.246	-.372	-.478	-.013	-.165	-.310	-.469	-.951
.350	-.055	-.122	-.225	-.305	-.397	-.078	-.195	-.299	-.386	-.811
.450	-.078	-.157	-.223	-.275	-.361	-.134	-.216	-.276	-.340	-.639
.550	-.084	-.141	-.195	-.239	-.329	-.125	-.185	-.222	-.275	-.505
.650	-.067	-.118	-.158	-.205	-.289	-.108	-.147	-.167	-.205	-.415
.750	-.054	-.087	-.119	-.154	-.256	-.064	-.081	-.093	-.128	-.339
.850	-.013	-.028	-.049	-.076	-.177	.004	-.001	-.005	-.030	-.233
.900	.017	.011	-.004	-.031	-.120	.040	.042	.042	.017	-.166
	Right side									
.025	-.099	-.019	-.019	.393	.779	-1.080	-.048	.380	.589	.717
.075	-.334	-.130	.031	.311	.530	-.467	-.121	.172	.366	.536
.150	-.346	-.147	.010	.224	.381	-.367	-.130	.077	.240	.394
.250	-.260	-.104	-.011	.161	.280	-.304	-.150	.008	.143	.269
.350	-.242	-.121	-.043	.081	.180	-.287	-.171	-.045	.066	.181
.450	-.231	-.148	-.069	.019	.096	-.264	-.183	-.089	-.004	.091
.550	-.190	-.125	-.057	-.001	.069	-.217	-.168	-.105	-.037	.041
.650	-.148	-.095	-.039	.007	.055	-.155	-.125	-.075	-.031	.021
.750	-.110	-.065	-.022	.007	.032	-.075	-.056	-.028	-.007	.011
.850	-.030	-.001	.024	.042	.044	.013	.028	.039	.042	.027
.900	.004	.025	.042	.049	.034	.039	.048	.051	.045	-.001
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.284	.659	.296	-.252	-.723	.314	.603	.170	-.236	-.659
.025	.331	-.138	-1.183	-1.104	-.729	.193	-.251	-.898	-.667	-.551
.075	.154	-.165	-.642	-1.052	-.723	-.020	-.230	-.684	-.645	-.543
.150	.042	-.180	-.431	-1.007	-.708	-.136	-.195	-.363	-.657	-.546
.250	-.043	-.191	-.328	-.911	-.685	-.098	-.121	-.246	-.604	-.556
.350	-.104	-.209	-.289	-.710	-.665					
.450	-.145	-.216	-.261	-.480	-.644					
.550	-.157	-.194	-.219	-.322	-.607					
.650	-.116	-.136	-.149	-.201	-.572	-.125	-.080	-.134	-.361	-.475
.750	-.057	-.062	-.070	-.105	-.528	-.069	-.019	-.087	-.323	-.449
.850	-.004	.001	-.005	-.033	-.469	-.025	.023	-.051	-.281	-.425
.900	.020	.031	.028	.036	-.443	-.004	.045	-.026	-.258	-.409
	Right side									
.025	-1.123	-.057	.384	.566	.664	-.999	-.139	.254	.401	.481
.075	-.581	-.136	.172	.360	.505	-.592	-.219	-.026	.130	.260
.150	-.389	-.145	.075	.233	.373	-.284	-.169	-.101	-.020	.093
.250	-.331	-.180	-.017	.117	.239					
.350	-.298	-.198	-.081	.024	.134					
.450	-.260	-.203	-.120	-.040	.047					
.550	-.211	-.180	-.122	-.070	-.012					
.650	-.143	-.122	-.090	-.061	-.035	-.105	-.062	-.098	-.125	-.175
.750	-.064	-.043	-.045	-.028	-.053	-.076	-.022	-.060	-.114	-.196
.850	.020	.037	.034	.025	-.052	-.017	.043	.004	-.069	-.190
.900	.058	.064	.057	.045	-.088	.011	.060	.020	-.070	-.213

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(h) $\alpha = 9.7°$; M = 0.85

$\frac{x}{C_v}$	C_p for — $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for — $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.339	.878	.424	-.240	-.021	.464	.739	-1.509	-1.009	-.655
.025	-.068	-.115	-.103	-.534	-1.479	.324	-.107	-1.180	-1.618	-1.064
.075	.013	-.132	-.308	-.558	-1.070	.149	-.154	-.468	-1.454	-1.083
.150	-.024	-.175	-.387	-.605	-.751	.055	-.161	-.381	-.711	-1.017
.250	-.032	-.121	-.266	-.434	-.508	-.019	-.178	-.328	-.478	-.932
.350	-.062	-.136	-.243	-.335	-.433	-.089	-.215	-.317	-.401	-.810
.450	-.089	-.168	-.239	-.290	-.393	-.148	-.246	-.298	-.385	-.673
.550	-.091	-.156	-.206	-.255	-.356	-.138	-.207	-.239	-.306	-.568
.650	-.074	-.131	-.169	-.216	-.328	-.118	-.162	-.176	-.226	-.485
.750	-.059	-.101	-.135	-.172	-.313	-.074	-.097	-.103	-.139	-.406
.850	-.018	-.036	-.059	-.087	-.226	.000	-.006	-.009	-.040	-.301
.900	.019	.011	-.012	-.039	-.170	.042	.044	.042	.011	-.237
	Right side									
.025	-.052	-.016	-.041	.325	.753	-1.069	-.050	.371	.586	.713
.075	-.332	-.135	.025	.289	.492	-.455	-.121	.166	.368	.526
.150	-.365	-.154	.005	.217	.347	-.372	-.139	.069	.236	.381
.250	-.271	-.108	-.022	.159	.252	-.315	-.159	-.004	.137	.259
.350	-.252	-.138	-.055	.077	.163	-.296	-.185	-.055	.061	.166
.450	-.241	-.159	-.085	.009	.078	-.271	-.209	-.106	-.020	.072
.550	-.191	-.135	-.072	-.014	.051	-.226	-.189	-.120	-.054	.019
.650	-.151	-.107	-.052	-.020	.029	-.162	-.139	-.092	-.050	-.008
.750	-.115	-.084	-.035	-.010	.016	-.075	-.065	-.043	-.023	-.017
.850	-.035	-.009	.018	.024	.019	.015	.023	.031	.031	-.008
.900	.005	.021	.036	.037	-.001	.048	.045	.046	.037	-.041

$\frac{x}{C_v}$	$z/b_v = 0.66$ $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$z/b_v = 0.93$ $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side									
.000	.336	.676	.368	-.139	-.656	.347	.605	.212	-.268	-.659
.025	.327	-.154	-1.251	-1.392	-.737	.194	-.266	-1.177	-.811	-.574
.075	.153	-.179	-.677	-1.296	-.739	-.034	-.290	-.719	-.785	-.563
.150	.036	-.199	-.461	-1.087	-.726	-.158	-.227	-.435	-.806	-.564
.250	-.051	-.216	-.354	-.838	-.703	-.117	-.141	-.246	-.705	-.578
.350	-.118	-.239	-.306	-.667	-.683					
.450	-.165	-.252	-.280	-.462	-.663					
.550	-.175	-.217	-.230	-.309	-.630					
.650	-.131	-.151	-.157	-.187	-.599	-.125	-.095	-.139	-.388	-.510
.750	-.067	-.075	-.078	-.097	-.558	-.065	-.030	-.088	-.331	-.485
.850	-.007	-.002	-.006	-.026	-.508	-.011	.021	-.051	-.278	-.455
.900	.023	.033	.032	.007	-.489	.000	.044	-.029	-.249	-.445
	Right side									
.025	-1.258	-.061	.378	.564	.664	-1.164	-.155	.246	.401	.483
.075	-.534	-.144	.166	.358	.503	-.701	-.260	-.043	.116	.261
.150	-.402	-.152	.069	.226	.364	-.310	-.192	-.130	-.052	.082
.250	-.349	-.195	-.029	.107	.232					
.350	-.310	-.227	-.103	.006	.117					
.450	-.268	-.232	-.142	-.063	.025					
.550	-.216	-.200	-.145	-.099	-.043	-.100	-.074	-.100	-.150	-.214
.650	-.145	-.136	-.109	-.083	-.070	-.071	-.033	-.065	-.127	-.232
.750	-.065	-.061	-.052	-.052	-.089	-.003	.043	.002	-.077	-.221
.850	.030	.033	.036	.017	-.086	.010	.061	.022	-.074	-.244
.900	.065	.062	.061	.033	-.123					

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(i) $\alpha = 9.7°$; M = 0.90

$\dfrac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.298	.908	.421	-.149	.109	.527	.750	.571	.147	-.478
.025	-.095	-.093	-.019	-.453	-1.313	.324	-.098	-1.041	-1.377	-.977
.075	.022	-.122	-.249	-.490	-.976	.153	-.154	-.523	-1.230	-.992
.150	-.005	-.178	-.411	-.587	-.728	.055	-.162	-.463	-.740	-.955
.250	-.024	-.121	-.289	-.505	-.606	-.018	-.181	-.337	-.645	-.883
.350	-.060	-.142	-.255	-.405	-.506	-.095	-.229	-.355	-.466	-.798
.450	-.095	-.178	-.269	-.322	-.437	-.164	-.272	-.345	-.308	-.682
.550	-.101	-.173	-.221	-.245	-.381	-.163	-.236	-.237	-.305	-.588
.650	-.089	-.146	-.171	-.210	-.339	-.144	-.181	-.181	-.253	-.521
.750	-.071	-.109	-.134	-.179	-.342	-.088	-.104	-.103	-.152	-.451
.850	-.020	-.033	-.062	-.094	-.281	-.004	-.003	-.003	-.040	-.354
.900	.015	.011	-.006	-.039	-.204	.040	.047	.049	.016	-.288
Right side										
.025	.039	.001	-.054	.320	.751	-.969	-.042	.375	.587	.724
.075	-.271	-.126	.034	.295	.503	-.568	-.121	.173	.368	.538
.150	-.408	-.158	.021	.230	.364	-.424	-.145	.074	.238	.394
.250	-.293	-.116	-.007	.166	.268	-.327	-.162	.002	.139	.273
.350	-.276	-.146	-.050	.080	.171	-.341	-.202	-.062	.057	.176
.450	-.274	-.174	-.079	.007	.082	-.310	-.233	-.119	-.027	.078
.550	-.208	-.153	-.077	-.013	.051	-.234	-.214	-.139	-.063	.016
.650	-.155	-.112	-.057	-.011	.030	-.168	-.160	-.114	-.064	-.013
.750	-.119	-.088	-.045	-.019	.006	-.080	-.069	-.053	-.031	-.025
.850	-.039	-.007	.011	.020	.003	.018	.023	.027	.024	-.025
.900	.006	.025	.033	.030	-.019	.052	.046	.043	.031	-.059

$\dfrac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.396	.688	.438	.006	-.532	.383	.607	.279	-.143	-.604
.025	.321	-.145	-1.179	-1.417	-.711	.178	-.274	-1.137	-.898	-.565
.075	.149	-.178	-.997	-1.311	-.721	-.056	-.331	-1.118	-.837	-.553
.150	.031	-.210	-.421	-1.130	-.717	-.262	-.248	-.511	-.828	-.553
.250	-.064	-.234	-.413	-.795	-.701	-.153	-.136	-.195	-.744	-.567
.350	-.144	-.281	-.378	-.639	-.684					
.450	-.214	-.285	-.249	-.523	-.658					
.550	-.220	-.246	-.233	-.397	-.630					
.650	-.151	-.158	-.157	-.253	-.593	-.137	-.093	-.147	-.412	-.513
.750	-.077	-.076	-.073	-.126	-.561	-.073	-.021	-.093	-.357	-.492
.850	-.009	.006	.005	-.028	-.515	-.021	.034	-.051	-.292	-.467
.900	.023	.039	.039	.011	-.498	-.002	.051	-.029	-.254	-.453
Right side										
.025	-1.156	-.062	.371	.561	.676	-1.184	-.145	.241	.403	.499
.075	-.938	-.144	.165	.350	.509	-1.061	-.332	-.065	.117	.273
.150	-.358	-.158	.066	.225	.372	-.401	-.212	-.202	-.070	.089
.250	-.409	-.210	-.038	.105	.242					
.350	-.371	-.253	-.123	-.001	.124					
.450	-.262	-.265	-.181	-.089	.024					
.550	-.235	-.228	-.179	-.129	-.047					
.650	-.155	-.150	-.127	-.112	-.082	-.115	-.077	-.113	-.160	-.235
.750	-.072	-.060	-.059	-.063	-.106	-.080	-.026	-.073	-.134	-.245
.850	.028	.041	.033	.012	-.094	-.012	.049	.007	-.077	-.226
.900	.064	.071	.059	.032	-.128	.011	.065	.026	-.070	-.245

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(j) $\alpha = 9.7°$; M = 0.92

$z/b_v = 0.11$ and $z/b_v = 0.38$

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
		$z/b_v = 0.11$					$z/b_v = 0.38$			
Left side										
.000	.308	.971	.451	-.123		.562	.742	.593	.175	
.025	-.065	-.053	.041	-.403		.330	-.049	-.989	-1.369	
.075	.040	-.077	-.166	-.447		.162	-.108	-.419	-1.227	
.150	.023	-.140	-.329	-.555		.068	-.130	-.372	-.734	
.250	.004	-.102	-.243	-.516		-.009	-.155	-.317	-.658	
.350	-.040	-.132	-.235	-.431		-.084	-.208	-.347	-.545	
.450	-.078	-.178	-.261	-.355		-.153	-.263	-.370	-.308	
.550	-.095	-.188	-.232	-.252		-.172	-.251	-.249	-.299	
.650	-.089	-.172	-.184	-.206		-.151	-.205	-.188	-.259	
.750	-.073	-.145	-.151	-.169		-.094	-.119	-.115	-.162	
.850	-.024	-.065	-.073	-.096		-.002	-.012	-.013	-.045	
.900	.010	-.008	-.015	-.043		.037	.042	.040	.010	
Right side										
.025	.114	.044	.004	.302		-.860	-.007	.393	.592	
.075	-.186	-.083	.066	.320		-.465	-.084	.193	.375	
.150	-.332	-.123	.059	.256		-.374	-.111	.096	.248	
.250	-.256	-.095	.033	.179		-.303	-.144	.018	.141	
.350	-.260	-.134	-.010	.092		-.337	-.189	-.052	.055	
.450	-.273	-.176	-.056	.012		-.337	-.231	-.113	-.027	
.550	-.216	-.170	-.069	-.009		-.236	-.226	-.147	-.070	
.650	-.160	-.134	-.060	-.009		-.173	-.176	-.122	-.079	
.750	-.126	-.116	-.048	-.020		-.084	-.087	-.061	-.043	
.850	-.042	-.027	.006	.018		.011	.015	.020	.016	
.900	-.005	.006	.027	.023		.048	.042	.034	.022	

$z/b_v = 0.66$ and $z/b_v = 0.93$

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
		$z/b_v = 0.66$					$z/b_v = 0.93$			
Left side										
.000	.413	.679	.438	.036		.411	.610	.295	-.108	
.025	.324	-.104	-1.126	-1.447		.186	-.245	-1.120	-.822	
.075	.157	-.142	-.930	-1.318		-.044	-.301	-1.077	-.816	
.150	.040	-.176	-.402	-1.153		-.249	-.243	-.507	-.826	
.250	-.057	-.214	-.409	-.813		-.152	-.134	-.192	-.784	
.350	-.141	-.280	-.430	-.640						
.450	-.215	-.300	-.258	-.542						
.550	-.219	-.266	-.231	-.431						
.650	-.156	-.171	-.167	-.279		-.137	-.102	-.150	-.431	
.750	-.078	-.078	-.081	-.148		-.066	-.031	-.095	-.370	
.850	-.006	-.002	-.003	-.045		-.017	.026	-.056	-.309	
.900	.024	.035	.034	-.002		.004	.047	-.031	-.277	
Right side										
.025	-1.080	-.035	.383	.556		-1.118	-.133	.249	.401	
.075	-.789	-.119	.181	.352		-.977	-.314	-.053	.115	
.150	-.361	-.137	.079	.227		-.363	-.210	-.193	-.067	
.250	-.396	-.199	-.028	.102						
.350	-.420	-.252	-.117	-.004						
.450	-.247	-.273	-.186	-.094						
.550	-.233	-.239	-.197	-.139						
.650	-.151	-.176	-.134	-.127		-.112	-.079	-.115	-.176	
.750	-.068	-.045	-.068	-.076		-.074	-.029	-.074	-.144	
.850	.029	.036	.027	.002		-.005	.049	.000	-.080	
.900	.069	.068	.053	.023		.015	.064	.020	-.082	

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0^\circ$ - Continued

(k) $\alpha = 15.6^\circ$;   M = 0.60

$\frac{x}{c_v}$	Cp for-					Cp for-				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	-.332	.815	.199	.103	-.514	.274	.733	.409	-.402	-.999
.025	-.644	-.158	.296	.112	-.532	.209	-.112	-.350	-.726	-1.090
.075	-.198	-.158	.021	-.089	-.462	.080	-.158	-.319	-.530	-.931
.150	-.085	-.178	-.141	-.206	-.437	.032	-.162	-.280	-.416	-.746
.250	-.042	-.139	-.152	-.195	-.371	-.011	-.169	-.265	-.361	-.600
.350	-.031	-.123	-.175	-.206	-.335	-.054	-.196	-.269	-.323	-.501
.450	-.038	-.142	-.193	-.208	-.297	-.081	-.208	-.253	-.296	-.424
.550	-.047	-.153	-.179	-.193	-.251	-.076	-.185	-.217	-.251	-.355
.650	-.045	-.121	-.143	-.157	-.210	-.072	-.158	-.179	-.206	-.297
.750	-.051	-.110	-.125	-.132	-.170	-.063	-.107	-.118	-.150	-.224
.850	-.035	-.066	-.087	-.089	-.108	-.020	-.046	-.062	-.093	-.154
.900	-.022	-.032	-.055	-.055	-.070	-.178	-.190	-.166	-.177	-.160
Right side										
.025	.324	-.103	-.798	-.298	-.102	-.569	-.110	.112	.197	.200
.075	-.072	-.185	-.229	-.089	.012	-.409	-.160	.024	.121	.214
.150	-.198	-.167	-.075	-.010	.087	-.332	-.162	.012	.089	.216
.250	-.194	-.121	-.015	.035	.132	-.293	-.167	.001	.080	.227
.350	-.203	-.130	-.001	.053	.148	-.271	-.178	-.024	.064	.205
.450	-.212	-.144	-.008	.044	.134	-.259	-.194	-.046	.042	.182
.550	-.189	-.139	-.017	.033	.100	-.219	-.180	-.053	.028	.146
.650	-.149	-.112	-.021	.017	.066	-.173	-.144	-.044	.024	.114
.750	-.119	-.094	-.017	.008	.041	-.112	-.091	-.024	.024	.087
.850	-.069	-.037	-.003	.015	.019	-.047	-.023	.008	.035	.069
.900	-.047	-.014	.003	.010	.012	-.020	-.012	.003	.021	.046
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	-.020	.630	.105	-.636	-.984	.098	.543	-.145	-.620	-.816
.025	.370	-.130	-.963	-1.328	-1.022	.200	-.208	-.949	-.810	-.650
.075	.189	-.160	-.544	-1.199	-1.002	.010	-.190	-.470	-.778	-.650
.150	.078	-.171	-.402	-.893	-.968	-.094	-.171	-.303	-.706	-.637
.250	.001	-.178	-.321	-.517	-.879	-.072	-.114	-.229	-.611	-.600
.350	-.054	-.194	-.287	-.355	-.741					
.450	-.087	-.201	-.262	-.283	-.598					
.550	-.099	-.190	-.226	-.240	-.473					
.650	-.085	-.149	-.172	-.177	-.369	-.110	-.085	-.147	-.343	-.451
.750	-.051	-.094	-.107	-.123	-.287	-.074	-.053	-.123	-.289	-.410
.850	-.015	-.037	-.051	-.075	-.215	-.054	-.014	-.096	-.235	-.369
.900	-.006	-.016	-.024	-.046	-.183	-.035	.004	-.069	-.202	-.342
Right side										
.025	-1.038	-.112	.382	.553	.558	-.811	-.167	.220	.334	.391
.075	-.599	-.162	.166	.348	.431	-.531	-.201	-.006	.100	.202
.150	-.395	-.160	.078	.224	.343	-.339	-.180	-.098	-.042	.055
.250	-.332	-.176	.010	.130	.254					
.350	-.305	-.194	-.042	.066	.191					
.450	-.268	-.201	-.071	.033	.141					
.550	-.228	-.176	-.078	.008	.105					
.650	-.169	-.137	-.055	.008	.084	-.081	-.089	-.084	-.084	-.086
.750	-.106	-.085	-.030	.017	.057	-.121	-.057	-.084	-.093	-.120
.850	-.024	-.005	.021	.044	.039	-.069	.004	-.024	-.053	-.111
.900	.001	.016	.033	.037	.021	-.047	.018	-.010	-.046	-.133

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Continued

(l) $\alpha = 15.8°$; M = 0.80

$z/b_v = 0.11$ and $z/b_v = 0.38$

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.11$					C_p for — $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	-.025	.857	.398	.216	-.494	.399	.774	.420	-.203	-.656
.025	-.991	.095	.351	.081	-.460	.004	-.054	-.357	-.600	-.795
.075	-.639	-.105	.005	-.115	-.429	-.078	-.155	-.355	-.502	-.720
.150	-.299	-.209	-.210	-.254	-.429	-.076	-.174	-.323	-.441	-.630
.250	-.144	-.152	-.193	-.219	-.372	-.066	-.187	-.304	-.405	-.561
.350	-.080	-.133	-.208	-.227	-.329	-.076	-.212	-.292	-.372	-.493
.450	-.046	-.147	-.219	-.224	-.287	-.083	-.231	-.278	-.330	-.433
.550	-.035	-.153	-.202	-.195	-.241	-.064	-.208	-.236	-.280	-.365
.650	-.028	-.115	-.157	-.163	-.202	-.055	-.174	-.193	-.233	-.307
.750	-.029	-.114	-.146	-.142	-.165	-.037	-.114	-.133	-.165	-.237
.850	-.022	-.068	-.099	-.089	-.106	-.006	-.041	-.063	-.100	-.170
.900	-.005	-.033	-.064	-.059	-.074	-.172	-.179	-.145	-.147	-.137
Right side										
.025	.420	-.772	-.961	-.517	-.293	-.465	-.293	-.036	.025	.044
.075	-.011	-.422	-.664	-.269	-.176	-.392	-.278	-.105	-.020	.085
.150	-.217	-.257	-.325	-.172	-.064	-.349	-.225	-.092	-.020	.141
.250	-.214	-.152	-.152	-.094	.035	-.310	-.176	-.064	.004	.185
.350	-.226	-.102	-.080	-.038	.096	-.293	-.165	-.057	.013	.199
.450	-.220	-.091	-.046	-.005	.125	-.272	-.174	-.063	.016	.191
.550	-.188	-.085	-.031	.012	.110	-.237	-.155	-.055	.018	.164
.650	-.148	-.068	-.020	.013	.075	-.185	-.117	-.042	.025	.135
.750	-.134	-.062	-.019	.016	.047	-.118	-.068	-.014	.032	.099
.850	-.073	-.024	-.002	.022	.029	-.046	-.003	.016	.044	.075
.900	-.055	-.007	.004	.021	.013	-.023	.008	.010	.029	.041

$z/b_v = 0.66$ and $z/b_v = 0.93$

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.320	.667	.430	.001	-.538	.241	.572	.128	-.348	-.745
.025	.312	-.109	-1.076	-1.314	-.901	.177	-.257	-1.064	-.833	-.628
.075	.143	-.159	-.572	-1.159	-.890	-.011	-.234	-.619	-.784	-.605
.150	.042	-.188	-.436	-.878	-.856	-.115	-.191	-.308	-.710	-.588
.250	-.012	-.202	-.334	-.544	-.792	-.064	-.114	-.223	-.610	-.563
.350	-.060	-.217	-.299	-.385	-.697					
.450	-.093	-.228	-.275	-.296	-.580					
.550	-.102	-.214	-.237	-.239	-.475					
.650	-.076	-.159	-.172	-.171	-.385	-.095	-.070	-.131	-.345	-.415
.750	-.034	-.085	-.099	-.104	-.305	-.064	-.026	-.095	-.281	-.382
.850	.000	-.024	-.028	-.050	-.240	-.031	.012	-.061	-.221	-.349
.900	.013	.002	.002	-.020	-.209	-.012	.034	-.036	-.181	-.330
Right side										
.025	-1.211	-.144	.325	.533	.621	-1.005	-.173	.201	.359	.456
.075	-.590	-.184	.124	.314	.462	-.647	-.226	-.040	.109	.256
.150	-.413	-.153	.051	.202	.367	-.331	-.178	-.125	-.050	.082
.250	-.349	-.168	-.011	.122	.291					
.350	-.310	-.184	-.061	.054	.222					
.450	-.272	-.184	-.083	.022	.164					
.550	-.226	-.162	-.078	.006	.127					
.650	-.162	-.115	-.057	.015	.097	-.115	-.064	-.067	-.071	-.090
.750	-.089	-.057	-.028	.024	.072	-.101	-.030	-.061	-.076	-.110
.850	-.003	.023	.036	.059	.055	-.034	.037	.002	-.026	-.101
.900	.027	.047	.048	.054	.024	-.015	.049	.017	-.024	-.124

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION; $i_t = 0°$ - Concluded

(m) $\alpha = 15.9°$; $M = 0.85$

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	-.357	.905	.565	.243	-.610	.441	.769	.463	-.211	-.589
.025	-1.038	-.009	.281	-.052	-.583	.244	-.183	-.520	-.672	-.764
.075	-.511	-.227	-.041	-.221	-.528	.030	-.267	-.476	-.576	-.699
.150	-.255	-.340	-.300	-.362	-.502	-.062	-.258	-.415	-.524	-.627
.250	-.200	-.237	-.260	-.300	-.438	-.114	-.243	-.362	-.485	-.568
.350	-.183	-.192	-.256	-.284	-.378	-.169	-.267	-.341	-.436	-.509
.450	-.183	-.196	-.252	-.269	-.315	-.204	-.286	-.315	-.381	-.442
.550	-.169	-.183	-.225	-.229	-.258	-.183	-.243	-.271	-.320	-.380
.650	-.132	-.136	-.169	-.187	-.216	-.159	-.200	-.222	-.260	-.323
.750	-.117	-.126	-.162	-.167	-.170	-.115	-.141	-.158	-.188	-.249
.850	-.080	-.079	-.114	-.111	-.112	-.053	-.059	-.085	-.116	-.190
.900	-.041	-.040	-.077	-.073	-.070	-.191	-.195	-.154	-.153	-.137
Right side										
.025	.234	-.687	-1.129	-.582	-.419	-1.084	-.317	-.033	-.039	-.021
.075	-.204	-.887	-.757	-.356	-.293	-.798	-.360	-.149	-.099	.011
.150	-.504	-.364	-.371	-.257	-.166	-.680	-.299	-.159	-.099	.062
.250	-.491	-.262	-.234	-.184	-.053	-.377	-.251	-.136	-.066	.119
.350	-.286	-.195	-.169	-.116	.026	-.307	-.234	-.125	-.053	.143
.450	-.252	-.161	-.117	-.068	.072	-.294	-.237	-.125	-.042	.146
.550	-.215	-.127	-.081	-.036	.085	-.256	-.203	-.104	-.025	.136
.650	-.160	-.096	-.057	-.019	.071	-.197	-.152	-.074	-.007	.119
.750	-.134	-.079	-.040	-.005	.052	-.121	-.096	-.044	.008	.093
.850	-.070	-.037	-.013	.008	.034	-.046	-.021	-.002	.025	.069
.900	-.048	-.024	-.007	.003	.022	-.008	-.011	-.010	.012	.042
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.297	.657	.452	.115	-.341	.255	.567	.179	-.294	-.687
.025	.289	-.172	-1.226	-1.427	-.858	.080	-.302	-1.234	-.983	-.580
.075	.109	-.217	-.685	-1.235	-.844	-.075	-.315	-.732	-.903	-.585
.150	-.013	-.240	-.497	-.873	-.820	-.228	-.237	-.377	-.794	-.576
.250	-.089	-.244	-.384	-.623	-.773	-.117	-.144	-.246	-.679	-.548
.350	-.155	-.261	-.324	-.473	-.693					
.450	-.198	-.272	-.298	-.361	-.590					
.550	-.204	-.253	-.263	-.270	-.489					
.650	-.160	-.185	-.189	-.191	-.395	-.131	-.090	-.148	-.356	-.418
.750	-.094	-.106	-.111	-.117	-.318	-.097	-.045	-.109	-.294	-.387
.850	-.042	-.040	-.037	-.056	-.253	-.051	.006	-.071	-.232	-.353
.900	-.015	-.007	-.007	-.026	-.220	-.022	.024	-.046	-.192	-.327
Right side										
.025	-1.253	-.186	.284	.506	.646	-1.256	-.234	.169	.349	.478
.075	-1.032	-.243	.077	.277	.469	-.866	-.308	-.090	.090	.270
.150	-.632	-.215	.001	.168	.367	-.506	-.231	-.188	-.094	.091
.250	-.301	-.229	-.061	.081	.283					
.350	-.289	-.243	-.117	.019	.213					
.450	-.276	-.234	-.134	-.019	.169					
.550	-.242	-.202	-.124	-.028	.129					
.650	-.176	-.147	-.094	-.014	.102	-.096	-.081	-.104	-.096	-.060
.750	-.104	-.082	-.051	.008	.073	-.110	-.048	-.084	-.092	-.110
.850	-.014	.011	.020	.044	.058	-.045	.028	-.011	-.038	-.099
.900	.024	.035	.034	.046	.024	-.017	.039	.001	-.036	-.123

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL TAIL

(a) $\alpha = 0°$; M = 0.60

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.642	.645	.338	-.425	-1.270	.279	.728	.329	-.416	-1.192
.025	.263	-.113	-.543	-1.130	-2.267	.330	-.107	-.623	-1.079	-1.133
.075	.123	-.120	-.369	-.669	-1.505	.135	-.143	-.431	-.939	-1.105
.150	.036	-.146	-.302	-.476	-.667	.050	-.146	-.318	-.697	-1.057
.250	.013	-.107	-.204	-.303	-.416	-.017	-.152	-.263	-.425	-1.005
.350	-.021	-.111	-.167	-.236	-.336	-.072	-.171	-.245	-.328	-.843
.450	-.067	-.129	-.156	-.199	-.281	-.104	-.175	-.222	-.266	-.639
.550	-.069	-.107	-.112	-.146	-.288	-.095	-.143	-.161	-.195	-.477
.650	-.040	-.065	-.062	-.093	-.210	-.067	-.100	-.108	-.137	-.352
.750	-.010	-.021	-.016	-.047	-.123	-.028	-.040	-.039	-.065	-.228
.850	.032	.029	.036	-.003	-.037	.036	.025	.029	.011	-.117
.900	.055	.059	.066	.031	.009	.064	.061	.068	.050	-.057
Right side										
.025	-.452	-.035	.306	.580	.815	-.620	-.063	.364	.591	.735
.075	-.374	-.118	.123	.333	.536	-.418	-.125	.148	.349	.516
.150	-.294	-.127	.043	.204	.374	-.317	-.134	.055	.209	.361
.250	-.205	-.104	.013	.130	.265	-.253	-.146	-.014	.105	.230
.350	-.179	-.120	-.035	.057	.164	-.226	-.159	-.062	.036	.146
.450	-.159	-.132	-.074	.001	.091	-.209	-.162	-.094	-.012	.071
.550	-.120	-.109	-.064	-.017	.057	-.166	-.141	-.090	-.036	.027
.650	-.065	-.061	-.039	.001	.050	-.104	-.100	-.064	-.024	.023
.750	-.012	-.017	-.009	.020	.052	-.037	-.031	-.019	.006	.027
.850	.052	.043	.045	.059	.080	.043	.038	.055	.054	.055
.900	.077	.064	.061	.066	.073	.064	.059	.064	.059	.039
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.240	.716	.279	-.388	-.660	.381	.668	.240	-.284	-.438
.025	.337	-.141	-.808	-1.054	-.726	.185	-.205	-.652	-.860	-.518
.075	.151	-.159	-.481	-.953	-.710	.016	-.164	-.371	-.803	-.518
.150	.036	-.169	-.360	-.780	-.696	-.046	-.148	-.273	-.701	-.553
.250	-.030	-.171	-.279	-.529	-.664	-.037	-.113	-.213	-.559	-.546
.350	-.081	-.182	-.247	-.360	-.633					
.450	-.111	-.185	-.222	-.275	-.601					
.550	-.111	-.162	-.181	-.211	-.555					
.650	-.083	-.118	-.117	-.151	-.500	-.069	-.070	-.096	-.365	-.391
.750	-.044	-.049	-.046	-.079	-.432	-.051	-.038	-.071	-.342	-.363
.850	.018	.006	.009	-.019	-.365	-.021	-.008	-.048	-.294	-.336
.900	.036	.034	.036	.008	-.327	-.001	.018	-.028	-.257	-.322
Right side										
.025	-.705	-.074	.380	.584	.683	-.551	-.141	.219	.368	.441
.075	-.466	-.141	.148	.345	.486	-.361	-.164	.009	.110	.208
.150	-.329	-.146	.055	.206	.329	-.264	-.155	-.046	.006	.082
.250	-.278	-.166	-.025	.093	.199					
.350	-.255	-.180	-.083	.013	.098					
.450	-.218	-.169	-.108	-.033	.032					
.550	-.170	-.150	-.110	-.052	-.005					
.650	-.108	-.107	-.074	-.040	-.016	-.070	-.065	-.058	-.068	-.080
.750	-.044	-.047	-.030	-.012	-.028	-.061	-.047	-.062	-.072	-.110
.850	.029	.025	.032	.036	-.025	-.010	.011	.000	-.024	-.107
.900	.059	.057	.064	.050	-.055	-.005	.025	.000	-.017	-.137

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(b) α = 0°; M = 0.80

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.745	.655	.520	.212	-.247	.479	.761	.548	.061	-.547
.025	.268	-.099	-.546	-1.127	-1.405	.319	-.099	-.935	-1.656	-.950
.075	.134	-.126	-.378	-.875	-1.191	.140	-.159	-.455	-1.469	-.962
.150	.035	-.171	-.333	-.567	-.988	.046	-.171	-.346	-.486	-.930
.250	.011	-.134	-.223	-.314	-.708	-.026	-.183	-.284	-.355	-.842
.350	-.033	-.143	-.187	-.241	-.486	-.085	-.205	-.269	-.323	-.738
.450	-.079	-.165	-.175	-.206	-.386	-.133	-.223	-.240	-.273	-.623
.550	-.085	-.143	-.127	-.144	-.293	-.118	-.182	-.175	-.188	-.513
.650	-.053	-.093	-.065	-.094	-.190	-.083	-.130	-.107	-.111	-.425
.750	-.018	-.040	-.007	-.036	-.132	-.030	-.053	-.030	-.029	-.326
.850	.032	.023	.053	.022	-.059	.041	.029	.050	.049	-.225
.900	.064	.055	.083	.060	-.009	.079	.066	.090	.085	-.161
Right side										
.025	-.412	-.026	.314	.581	.815	-.921	-.099	.361	.587	.743
.075	-.396	-.131	.131	.344	.553	-.427	-.156	.152	.354	.532
.150	-.311	-.146	.048	.212	.382	-.336	-.160	.053	.213	.377
.250	-.220	-.134	.013	.142	.277	-.272	-.176	-.017	.113	.254
.350	-.190	-.146	-.040	.060	.170	-.249	-.197	-.074	.032	.152
.450	-.173	-.165	-.088	-.007	.081	-.223	-.205	-.117	-.029	.057
.550	-.123	-.134	-.088	-.026	.038	-.171	-.174	-.118	-.056	.008
.650	-.057	-.082	-.052	-.012	.028	-.094	-.120	-.077	-.036	-.004
.750	.002	-.031	-.017	.009	.025	-.013	-.036	-.016	.005	.002
.850	.064	.040	.047	.060	.052	.064	.044	.061	.063	.022
.900	.090	.061	.067	.069	.055	.096	.061	.076	.072	.003
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.444	.743	.491	.020	-.390	.477	.689	.384	-.095	-.287
.025	.332	-.136	-1.098	-1.626	-.714	.142	-.236	-1.052	-1.237	-.494
.075	.149	-.177	-.555	-1.320	-.711	-.009	-.210	-.486	-1.038	-.469
.150	.025	-.199	-.400	-.978	-.714	-.077	-.186	-.283	-.805	-.504
.250	-.048	-.208	-.306	-.655	-.671	-.047	-.139	-.234	-.592	-.506
.350	-.106	-.223	-.263	-.395	-.633					
.450	-.141	-.226	-.234	-.247	-.582					
.550	-.130	-.197	-.185	-.170	-.521					
.650	-.091	-.131	-.113	-.095	-.472	-.062	-.077	-.089	-.320	-.387
.750	-.042	-.057	-.034	-.032	-.415	-.045	-.040	-.055	-.293	-.357
.850	.034	.012	.033	.025	-.355	-.006	.006	-.028	-.249	-.333
.900	.057	.041	.064	.052	-.323	.020	.029	-.010	-.200	-.317
Right side										
.025	-1.179	-.105	.377	.588	.710	-1.073	-.200	.201	.368	.479
.075	-.501	-.180	.152	.350	.508	-.421	-.216	-.023	.090	.225
.150	-.351	-.176	.053	.212	.356	-.266	-.188	-.068	-.023	.081
.250	-.299	-.203	-.031	.091	.213					
.350	-.264	-.226	-.104	-.004	.102					
.450	-.223	-.216	-.132	-.054	.025					
.550	-.168	-.186	-.129	-.076	-.026					
.650	-.097	-.125	-.089	-.050	-.042	-.061	-.063	-.055	-.056	-.083
.750	-.023	-.050	-.030	-.015	-.050	-.042	-.043	-.045	-.056	-.123
.850	.061	.040	.053	.050	-.044	.020	.027	.021	-.001	-.109
.900	.095	.069	.079	.069	-.070	.026	.043	.033	-.009	-.140

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(c) α = 0°; M = 0.85

$\frac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.772	-.676	-.578	-.331	-.059	.529	-.776	-.612	-.213	-.343
.025	.270	-.070	-.525	-1.034	-1.166	.322	-.078	-1.062	-1.446	-.861
.075	.136	-.103	-.381	-.824	-1.025	.139	-.142	-.434	-1.290	-.868
.150	.034	-.152	-.353	-.676	-.895	.046	-.159	-.357	-1.146	-.861
.250	.004	-.122	-.236	-.374	-.729	-.031	-.176	-.304	-.368	-.773
.350	-.043	-.136	-.199	-.266	-.575	-.100	-.206	-.281	-.288	-.711
.450	-.096	-.162	-.187	-.203	-.467	-.158	-.226	-.253	-.223	-.617
.550	-.101	-.136	-.132	-.135	-.357	-.141	-.183	-.177	-.151	-.525
.650	-.067	-.085	-.069	-.084	-.251	-.097	-.125	-.107	-.087	-.449
.750	-.030	-.029	-.002	-.023	-.179	-.047	-.042	-.029	-.013	-.359
.850	.030	.041	.063	.037	-.094	.042	.045	.059	.068	-.268
.900	.062	.078	.092	.072	-.036	.083	.091	.099	.105	-.202
Right side										
.025	-.386	-.013	.316	.584	.814	-.950	-.089	.357	.582	.743
.075	-.400	-.119	.130	.345	.555	-.434	-.147	.149	.351	.530
.150	-.328	-.140	.049	.216	.388	-.351	-.159	.051	.216	.376
.250	-.237	-.127	.012	.143	.279	-.291	-.176	-.025	.111	.248
.350	-.206	-.145	-.046	.055	.172	-.266	-.199	-.088	.023	.145
.450	-.186	-.163	-.100	-.020	.070	-.234	-.206	-.135	-.050	.043
.550	-.126	-.136	-.106	-.043	.023	-.174	-.176	-.137	-.074	-.013
.650	-.060	-.076	-.063	-.024	.009	-.094	-.115	-.095	-.054	-.026
.750	.002	-.025	-.025	.000	.004	-.011	-.023	-.023	-.003	-.018
.850	.073	.053	.046	.057	.033	.074	.063	.062	.064	-.001
.900	.096	.079	.068	.069	.039	.099	.082	.076	.079	-.020
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.492	.761	.555	.169	-.252	.500	.699	.430	.037	-.226
.025	.334	-.115	-1.153	-1.541	-.702	.137	-.234	-1.146	-1.374	-.497
.075	.150	-.163	-.629	-1.382	-.698	-.037	-.216	-.667	-1.124	-.479
.150	.026	-.192	-.417	-1.234	-.712	-.103	-.184	-.249	-.990	-.517
.250	-.061	-.204	-.313	-.634	-.650	-.063	-.130	-.241	-.611	-.513
.350	-.127	-.226	-.273	-.226	-.624					
.450	-.164	-.230	-.247	-.189	-.583					
.550	-.157	-.193	-.189	-.152	-.530					
.650	-.103	-.122	-.109	-.091	-.479	-.071	-.065	-.092	-.323	-.404
.750	-.054	-.039	-.033	-.027	-.421	-.044	-.022	-.059	-.294	-.378
.850	.032	.034	.038	.034	-.360	-.003	.028	-.028	-.253	-.346
.900	.054	.061	.069	.065	-.330	.020	.053	-.012	-.205	-.329
Right side										
.025	-1.144	-.100	.370	.578	.711	-1.140	-.207	.183	.356	.482
.075	-.460	-.172	.147	.346	.509	-.616	-.230	-.059	.068	.223
.150	-.373	-.170	.045	.212	.353	-.217	-.187	-.100	-.070	.074
.250	-.311	-.204	-.051	.087	.213					
.350	-.276	-.230	-.126	-.023	.095					
.450	-.234	-.219	-.157	-.075	.008					
.550	-.178	-.183	-.147	-.095	-.047					
.650	-.101	-.109	-.098	-.065	-.062	-.074	-.055	-.060	-.058	-.100
.750	-.023	-.029	-.035	-.021	-.070	-.050	-.025	-.049	-.056	-.139
.850	.070	.062	.056	.060	-.054	.020	.048	.025	.003	-.118
.900	.097	.093	.083	.078	-.081	.023	.062	.034	.006	-.149

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(d) $\alpha = 0°$; M = 0.90

$\frac{x}{C_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.801	.685	.616	.418	.076	.578	.792	.664	.316	-.200
.025	.278	-.064	-.495	-.981	-1.063	.319	-.086	-.931	-1.300	-.781
.075	.147	-.096	-.399	-.785	-.935	.140	-.153	-.677	-1.182	-.798
.150	.035	-.160	-.395	-.690	-.825	.043	-.172	-.427	-1.085	-.776
.250	.006	-.129	-.274	-.548	-.691	-.032	-.185	-.338	-.668	-.722
.350	-.046	-.147	-.232	-.362	-.585	-.112	-.223	-.356	-.492	-.679
.450	-.114	-.180	-.227	-.284	-.515	-.181	-.259	-.290	-.281	-.622
.550	-.123	-.153	-.154	-.161	-.432	-.168	-.204	-.189	-.122	-.555
.650	-.087	-.088	-.076	-.082	-.339	-.118	-.131	-.112	-.066	-.491
.750	-.038	-.026	-.010	-.020	-.250	-.045	-.036	-.030	-.004	-.411
.850	.029	.049	.054	.035	-.159	.041	.056	.056	.071	-.335
.900	.064	.085	.086	.069	-.100	.090	.105	.105	.109	-.281
	Right side									
.025	-.353	.013	.311	.584	.821	-.899	-.049	.334	.575	.745
.075	-.424	-.100	.122	.343	.561	-.632	-.137	.129	.340	.528
.150	-.362	-.137	.038	.209	.391	-.392	-.156	.032	.201	.375
.250	-.262	-.126	-.008	.136	.281	-.314	-.180	-.046	.092	.248
.350	-.231	-.150	-.068	.042	.170	-.311	-.215	-.117	.000	.139
.450	-.208	-.177	-.128	-.040	.060	-.246	-.232	-.181	-.082	.030
.550	-.135	-.147	-.140	-.075	.005	-.176	-.193	-.183	-.115	-.034
.650	-.060	-.082	-.096	-.052	-.010	-.091	-.117	-.131	-.090	-.057
.750	.008	-.021	-.049	-.032	-.028	.000	-.016	-.042	-.032	-.053
.850	.075	.060	.034	.041	.001	.085	.073	.051	.048	-.035
.900	.102	.089	.055	.056	-.007	.112	.094	.069	.060	-.067
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.542	.777	.614	.271	-.129	.533	.711	.478	.144	-.160
.025	.329	-.123	-1.045	-1.385	-.677	.120	-.307	-.993	-1.341	-.461
.075	.147	-.174	-.883	-1.255	-.681	-.072	-.276	-.965	-1.285	-.464
.150	.020	-.208	-.481	-1.165	-.696	-.179	-.208	-.489	-1.122	-.499
.250	-.075	-.232	-.414	-1.109	-.629	-.075	-.142	-.179	-.979	-.512
.350	-.153	-.259	-.287	-.676	-.610					
.450	-.199	-.258	-.235	-.284	-.575					
.550	-.179	-.209	-.201	-.093	-.534					
.650	-.123	-.118	-.116	-.032	-.496	-.069	-.057	-.100	-.241	-.425
.750	-.044	-.028	-.031	.004	-.456	-.034	-.014	-.059	-.260	-.400
.850	.036	.049	.043	.048	-.404	.010	.039	-.034	-.241	-.373
.900	.063	.078	.081	.069	-.380	.033	.065	-.011	-.201	-.362
	Right side									
.025	-1.033	-.065	.348	.563	.710	-1.068	-.199	.152	.337	.485
.075	-.833	-.158	.124	.330	.502	-.915	-.270	-.116	.038	.221
.150	-.366	-.165	.024	.199	.352	-.278	-.208	-.202	-.124	.063
.250	-.390	-.221	-.081	.067	.209					
.350	-.239	-.254	-.179	-.061	.084					
.450	-.234	-.247	-.219	-.138	-.019					
.550	-.180	-.196	-.197	-.157	-.086					
.650	-.092	-.111	-.130	-.111	-.100	-.063	-.048	-.070	-.084	-.132
.750	-.010	-.020	-.049	-.050	-.102	-.040	-.013	-.057	-.073	-.171
.850	.082	.076	.050	.041	-.077	.031	.065	.021	-.006	-.141
.900	.114	.107	.082	.063	-.114	.037	.077	.034	-.011	-.180

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(e) $\alpha = 0°$; $M = 0.92$

$\frac{x}{c_v}$	C_p for —					C_p for —				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.813	-.684	-.631	-.448	.112	.600	.796	.682	.357	-.160
.025	.272	-.059	-.471	-.928	-1.031	.311	-.081	-.916	-1.233	-.733
.075	.140	-.101	-.387	-.738	-.896	.131	-.159	-.710	-1.125	-.746
.150	.028	-.167	-.391	-.664	-.785	.034	-.177	-.432	-1.034	-.735
.250	-.005	-.139	-.289	-.566	-.660	-.045	-.197	-.340	-.727	-.700
.350	-.056	-.160	-.232	-.382	-.568	-.126	-.246	-.358	-.548	-.659
.450	-.127	-.204	-.231	-.316	-.514	-.207	-.297	-.334	-.415	-.601
.550	-.142	-.180	-.150	-.175	-.440	-.196	-.230	-.176	-.110	-.539
.650	-.102	-.104	-.066	-.073	-.358	-.135	-.133	-.098	-.031	-.478
.750	-.049	-.031	.002	-.005	-.268	-.056	-.037	-.017	.023	-.407
.850	.023	.049	.064	.050	-.173	.036	.057	.068	.090	-.338
.900	.058	.087	.096	.085	-.111	.085	.104	.110	.123	-.277
Right side										
.025	-.335	.005	.323	.591	.832	-.841	-.068	.346	.583	.754
.075	-.420	-.113	.135	.352	.568	-.656	-.151	.141	.348	.539
.150	-.380	-.142	.049	.221	.405	-.432	-.167	.039	.215	.385
.250	-.289	-.138	.002	.145	.290	-.326	-.193	-.046	.104	.256
.350	-.246	-.168	-.061	.052	.182	-.338	-.234	-.115	.009	.150
.450	-.225	-.198	-.123	-.032	.074	-.293	-.273	-.184	-.081	.037
.550	-.144	-.176	-.144	-.072	.012	-.171	-.217	-.191	-.118	-.029
.650	-.060	-.097	-.095	-.054	-.008	-.089	-.122	-.131	-.094	-.056
.750	.003	-.026	-.041	-.028	-.028	-.003	-.014	-.033	-.024	-.053
.850	.074	.055	.041	.050	-.002	.085	.074	.061	.059	-.039
.900	.097	.087	.061	.068	-.012	.110	.096	.076	.075	-.064
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.559	.778	.631	.314	-.102	.547	.712	.498	.191	-.140
.025	.317	-.120	-1.022	-1.312	-.677	.090	-.297	-.997	-1.282	-.488
.075	.140	-.177	-.877	-1.186	-.677	-.085	-.306	-.955	-1.224	-.454
.150	.008	-.217	-.674	-1.112	-.702	-.233	-.258	-.740	-1.076	-.486
.250	-.090	-.252	-.399	-1.072	-.616	-.122	-.151	-.199	-.949	-.502
.350	-.184	-.318	-.404	-.962	-.594					
.450	-.238	-.292	-.186	-.371	-.565					
.550	-.203	-.231	-.176	-.064	-.525					
.650	-.142	-.117	-.104	.002	-.494	-.074	-.059	-.094	-.289	-.417
.750	-.036	-.026	-.020	.038	-.456	-.041	-.009	-.057	-.269	-.397
.850	.034	.050	.055	.077	-.409	.005	.042	-.030	-.237	-.371
.900	.061	.082	.089	.098	-.379	.027	.071	-.012	-.203	-.357
Right side										
.025	-.971	-.083	.354	.571	.716	-1.013	-.217	.159	.348	.490
.075	-.820	-.175	.134	.341	.510	-.914	-.337	-.112	.053	.230
.150	-.543	-.180	.031	.207	.357	-.616	-.252	-.223	-.114	.070
.250	-.393	-.247	-.080	.069	.217					
.350	-.376	-.321	-.186	-.057	.085					
.450	-.193	-.279	-.232	-.142	-.015					
.550	-.173	-.218	-.201	-.167	-.083					
.650	-.098	-.109	-.124	-.112	-.102	-.069	-.048	-.064	-.068	-.122
.750	-.016	-.022	-.042	-.040	-.106	-.048	-.016	-.051	-.057	-.168
.850	.081	.078	.061	.056	-.077	.027	.065	.033	.010	-.136
.900	.111	.109	.089	.085	-.105	.028	.083	.043	.009	-.177

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(f) $\alpha = 9°$; M = 0.60

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.419	.828	.555	.172	-1.068	.075	.650	.007	-.980	-1.806
.025	-.142	-.122	-.023	-.426	-.993	.322	-.075	-.692	-1.297	-1.638
.075	.001	-.118	-.155	-.389	-.694	.159	-.093	-.386	-1.364	-2.051
.150	.014	-.136	-.196	-.352	-.514	.078	-.091	-.267	-.385	-1.631
.250	.012	-.104	-.139	-.251	-.346	.017	-.095	-.219	-.294	-.397
.350	-.013	-.100	-.139	-.207	-.285	-.029	-.120	-.208	-.271	-.355
.450	-.035	-.111	-.139	-.186	-.248	-.065	-.127	-.180	-.234	-.322
.550	-.047	-.104	-.119	-.151	-.210	-.054	-.093	-.139	-.179	-.259
.650	-.022	-.063	-.073	-.098	-.154	-.033	-.063	-.094	-.131	-.213
.750	-.008	-.036	-.037	-.040	-.098	-.002	-.018	-.039	-.075	-.161
.850	.023	.009	.014	.024	-.007	.044	.053	.027	-.001	-.079
.900	.055	.043	.041	.059	.040	.064	.089	.059	.033	-.030
Right side										
.025	.048	-.063	-.062	.193	.678	-.563	.032	.370	.530	.624
.075	-.158	-.116	.009	.172	.516	-.354	-.034	.185	.348	.481
.150	-.189	-.104	.030	.144	.390	-.262	-.052	.096	.223	.360
.250	-.149	-.084	.027	.156	.308	-.216	-.068	.036	.133	.241
.350	-.146	-.093	.000	.121	.217	-.192	-.084	-.005	.075	.175
.450	-.146	-.109	-.028	.080	.152	-.173	-.093	-.041	.024	.101
.550	-.121	-.097	-.032	.052	.115	-.140	-.082	-.043	.003	.068
.650	-.074	-.063	-.012	.047	.101	-.087	-.052	-.025	.006	.054
.750	-.042	-.027	.007	.052	.096	-.029	.009	.011	.029	.061
.850	.028	.023	.048	.091	.115	.035	.068	.062	.070	.082
.900	.046	.043	.064	.093	.119	.062	.091	.071	.073	.068
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.028	.582	-.059	-.736	-1.248	.182	.523	-.066	-.410	-.790
.025	.297	-.122	-.886	-1.057	-1.000	.143	-.204	-.731	-.629	-.603
.075	.141	-.129	-.441	-1.013	-.998	-.017	-.166	-.384	-.602	-.596
.150	.044	-.129	-.320	-.969	-1.002	-.094	-.132	-.235	-.551	-.577
.250	-.020	-.122	-.242	-.713	-1.026	-.065	-.070	-.149	-.458	-.547
.350	-.065	-.129	-.217	-.338	-.923					
.450	-.087	-.129	-.190	-.191	-.741					
.550	-.092	-.118	-.153	-.165	-.561					
.650	-.067	-.070	-.098	-.112	-.374	-.072	-.020	-.073	-.276	-.463
.750	-.026	-.009	-.034	-.050	-.208	-.035	.009	-.041	-.237	-.453
.850	.010	.028	.014	-.003	-.110	-.006	.043	-.007	-.186	-.425
.900	.028	.059	.041	.024	-.065	.010	.064	.009	-.158	-.399
Right side										
.025	-.671	.000	.354	.502	.575	-.542	-.084	.187	.278	.343
.075	-.404	-.066	.162	.322	.444	-.339	-.134	-.007	.066	.157
.150	-.280	-.077	.073	.202	.318	-.221	-.122	-.089	-.059	.000
.250	-.237	-.107	.000	.098	.203					
.350	-.214	-.111	-.053	.027	.117					
.450	-.183	-.118	-.078	-.013	.054					
.550	-.151	-.102	-.080	-.036	.016					
.650	-.097	-.066	-.055	-.027	.000	-.031	-.020	-.053	-.092	-.105
.750	-.033	-.013	-.018	-.006	.009	-.038	.003	-.043	-.087	-.131
.850	.030	.06	.052	.043	.035	.010	.055	.016	-.036	-.112
.900	.055	.084	.071	.057	.042	.023	.078	.030	-.031	-.128

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(g) $\alpha = 9°$; $M = 0.80$

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.426	.880	.576	.331	-.502	.302	.676	.270	-.358	-1.146
.025	-.178	-.123	.050	-.338	-.729	.314	-.074	-.804	-1.303	-1.475
.075	-.008	-.126	-.127	-.381	-.674	.155	-.098	-.407	-1.248	-1.475
.150	.010	-.152	-.218	-.399	-.606	.077	-.095	-.272	-.607	-1.295
.250	.011	-.118	-.157	-.286	-.379	.016	-.109	-.227	-.306	-.808
.350	-.016	-.109	-.151	-.228	-.323	-.036	-.139	-.220	-.283	-.521
.450	-.050	-.127	-.156	-.203	-.300	-.074	-.150	-.195	-.245	-.417
.550	-.060	-.129	-.137	-.163	-.251	-.065	-.115	-.145	-.189	-.338
.650	-.033	-.071	-.078	-.101	-.184	-.045	-.071	-.087	-.136	-.272
.750	-.008	-.043	-.031	-.039	-.120	-.008	-.019	-.026	-.067	-.193
.850	.033	.010	.027	.034	-.020	.057	.058	.049	.013	-.076
.900	.062	.042	.064	.077	.044	.086	.096	.088	.053	-.017
	Right side									
.025	.128	-.063	-.084	.180	.530	-.657	.021	.384	.540	.639
.075	-.137	-.133	.013	.180	.495	-.349	-.048	.193	.351	.499
.150	-.200	-.118	.045	.143	.411	-.262	-.060	.110	.238	.374
.250	-.158	-.097	.039	.135	.333	-.219	-.083	.049	.152	.266
.350	-.161	-.101	.007	.111	.236	-.203	-.103	.004	.085	.190
.450	-.163	-.120	-.026	.076	.160	-.192	-.118	-.043	.023	.110
.550	-.137	-.115	-.034	.053	.117	-.148	-.101	-.051	-.001	.069
.650	-.080	-.069	-.012	.053	.099	-.085	-.063	-.025	.005	.055
.750	-.036	-.039	.009	.053	.091	-.016	.007	.018	.033	.061
.850	.033	.024	.064	.095	.119	.060	.079	.081	.080	.084
.900	.051	.042	.082	.095	.120	.088	.090	.093	.083	.072
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.229	.601	.178	-.364	-.913	.273	.531	.120	-.245	-.770
.025	.290	-.127	-.936	-1.006	-.881	.131	-.242	-.698	-.571	-.577
.075	.137	-.135	-.549	-.983	-.906	-.037	-.155	-.537	-.543	-.578
.150	.041	-.147	-.331	-.977	-.871	-.114	-.155	-.302	-.503	-.572
.250	-.028	-.146	-.261	-.851	-.837	-.065	-.088	-.169	-.431	-.542
.350	-.071	-.152	-.223	-.548	-.793					
.450	-.103	-.158	-.192	-.182	-.717					
.550	-.109	-.139	-.153	-.136	-.613					
.650	-.070	-.080	-.083	-.099	-.481	-.054	-.011	-.051	-.306	-.414
.750	-.019	-.008	-.015	-.039	-.344	-.015	.027	-.017	-.275	-.396
.850	.033	.042	.042	.017	-.225	.024	.067	.023	-.231	-.368
.900	.056	.073	.073	.050	-.167	.047	.085	.041	-.197	-.356
	Right side									
.025	-.810	-.004	.355	.515	.588	-.722	-.110	.186	.310	.384
.075	-.418	-.078	.170	.331	.466	-.386	-.179	-.029	.091	.202
.150	-.279	-.088	.084	.215	.339	-.215	-.147	-.113	-.055	.018
.250	-.248	-.124	.004	.115	.222					
.350	-.226	-.139	-.055	.031	.123					
.450	-.187	-.146	-.083	-.018	.056					
.550	-.146	-.120	-.078	-.044	.011					
.650	-.083	-.078	-.047	-.035	.005	-.024	-.013	-.034	-.070	-.113
.750	-.018	.006	-.003	.002	-.001	-.010	.015	-.022	-.078	-.137
.850	.067	.079	.070	.065	.024	.050	.080	.042	-.032	-.119
.900	.091	.100	.090	.083	.017	.067	.096	.058	-.026	-.140

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL TAIL - Continued

(h) $\alpha = 9°$; $M = 0.85$

$\frac{x}{C_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.442	.892	.583	.367	-.428	.329	.677	.338	-.212	-.920
.025	-.176	-.130	.071	-.300	-.612	.322	-.074	-.821	-1.270	-1.435
.075	-.009	-.127	-.122	-.372	-.621	.164	-.103	-.435	-1.110	-1.295
.150	.016	-.153	-.239	-.432	-.637	.081	-.098	-.283	-.683	-1.140
.250	.016	-.123	-.172	-.314	-.408	.019	-.113	-.247	-.380	-.867
.350	-.019	-.114	-.167	-.254	-.328	-.044	-.150	-.240	-.311	-.585
.450	-.054	-.140	-.176	-.222	-.332	-.084	-.163	-.216	-.270	-.469
.550	-.063	-.140	-.155	-.181	-.283	-.077	-.126	-.156	-.212	-.385
.650	-.040	-.083	-.088	-.113	-.216	-.054	-.084	-.098	-.149	-.323
.750	-.013	-.051	-.038	-.050	-.143	-.013	-.024	-.031	-.076	-.242
.850	.034	.007	.024	.032	-.053	.057	.056	.051	.012	-.115
.900	.064	.044	.059	.075	.021	.090	.099	.095	.058	-.047
Right side										
.025	.143	-.058	-.107	.166	.451	-.791	-.001	.375	.534	.647
.075	-.135	-.141	-.001	.170	.466	-.371	-.063	.192	.345	.502
.150	-.216	-.126	.038	.194	.418	-.280	-.074	.103	.235	.379
.250	-.170	-.108	.034	.126	.340	-.235	-.094	.041	.154	.269
.350	-.177	-.114	.001	.101	.242	-.218	-.116	-.009	.078	.193
.450	-.175	-.131	-.041	.067	.162	-.200	-.134	-.059	.012	.109
.550	-.143	-.126	-.046	.047	.117	-.150	-.118	-.065	-.014	.061
.650	-.084	-.080	-.028	.042	.097	-.087	-.076	-.039	-.005	.044
.750	-.039	-.047	.001	.044	.084	-.013	.002	.011	.026	.050
.850	.034	.022	.055	.087	.109	.061	.079	.079	.077	.071
.900	.060	.046	.075	.094	.107	.096	.090	.091	.081	.057
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.247	.599	.240	-.285	-.804	.280	.530	.143	-.245	-.704
.025	.296	-.123	-.962	-1.136	-.847	.127	-.227	-.783	-.626	-.569
.075	.147	-.137	-.619	-1.107	-.841	-.040	-.238	-.595	-.599	-.568
.150	.046	-.151	-.351	-1.038	-.806	-.136	-.168	-.321	-.566	-.567
.250	-.026	-.157	-.274	-.813	-.767	-.074	-.097	-.167	-.485	-.552
.350	-.080	-.166	-.231	-.426	-.737					
.450	-.116	-.170	-.200	-.202	-.681					
.550	-.119	-.148	-.156	-.175	-.607					
.650	-.076	-.088	-.086	-.110	-.515	-.052	-.017	-.052	-.317	-.432
.750	-.017	-.011	-.016	-.037	-.408	-.012	.022	-.012	-.272	-.402
.850	.037	.044	.048	.025	-.303	.036	.067	.031	-.215	-.365
.900	.067	.076	.081	.057	-.248	.054	.089	.049	-.175	-.351
Right side										
.025	-.903	-.023	.353	.509	.591	-.844	-.133	.182	.315	.392
.075	-.485	-.097	.166	.328	.463	-.472	-.208	-.043	.087	.206
.150	-.296	-.103	.079	.213	.338	-.220	-.168	-.137	-.084	.011
.250	-.263	-.138	-.008	.104	.225					
.350	-.235	-.156	-.068	.019	.123					
.450	-.193	-.160	-.099	-.038	.046					
.550	-.149	-.131	-.096	-.063	.000					
.650	-.079	-.083	-.059	-.045	-.019	-.034	-.016	-.036	-.081	-.126
.750	-.012	.013	-.011	-.012	-.027	-.007	.016	-.019	-.083	-.157
.850	.073	.082	.071	.055	-.003	.056	.084	.048	-.027	-.139
.900	.104	.109	.095	.074	-.014	.074	.099	.065	-.018	-.157

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL TAIL - Continued

(i) $\alpha = 9°$; M = 0.90

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.462	.918	.602	.408	-.211	.391	.691	.409	-.056	-.707
.025	-.189	-.123	.104	-.230	-.493	.321	-.071	-1.007	-1.368	-1.325
.075	-.005	-.125	-.106	-.315	-.560	.169	-.106	-.429	-.890	-1.188
.150	.020	-.157	-.259	-.446	-.627	.086	-.109	-.322	-.671	-1.054
.250	.017	-.129	-.191	-.357	-.516	.023	-.125	-.270	-.518	-.866
.350	-.016	-.123	-.188	-.296	-.351	-.044	-.162	-.272	-.357	-.613
.450	-.054	-.154	-.200	-.254	-.358	-.092	-.191	-.242	-.309	-.508
.550	-.067	-.157	-.177	-.197	-.335	-.087	-.147	-.168	-.238	-.433
.650	-.043	-.093	-.098	-.121	-.263	-.064	-.094	-.101	-.155	-.385
.750	-.012	-.053	-.043	-.047	-.192	-.015	-.021	-.027	-.071	-.312
.850	.041	.009	.022	.037	-.107	.063	.067	.055	.021	-.188
.900	.074	.049	.061	.088	-.041	.098	.113	.102	.070	-.113
	Right side									
.025	.174	-.050	-.117	.151	.292	-.842	.005	.374	.531	.643
.075	-.112	-.143	-.007	.163	.398	-.372	-.062	.190	.349	.488
.150	-.226	-.127	.035	.141	.395	-.296	-.077	.103	.238	.364
.250	-.181	-.111	.029	.119	.331	-.246	-.098	.037	.149	.255
.350	-.190	-.119	-.007	.091	.230	-.242	-.129	-.015	.080	.173
.450	-.190	-.141	-.046	.058	.143	-.220	-.151	-.069	.001	.082
.550	-.158	-.138	-.058	.033	.089	-.159	-.134	-.081	-.029	.030
.650	-.086	-.087	-.037	.032	.063	-.088	-.085	-.051	-.025	.008
.750	-.036	-.050	-.010	.036	.042	-.007	.001	.005	.015	.011
.850	.043	.025	.055	.083	.070	.075	.087	.079	.071	.029
.900	.067	.051	.076	.095	.067	.108	.106	.092	.083	.014

$\frac{x}{c_v}$	$z/b_v = 0.66$ $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$z/b_v = 0.93$ $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side									
.000	.293	.614	.305	-.154	-.669	.301	.537	.174	-.257	-.629
.025	.294	-.122	-1.059	-1.412	-.831	.138	-.271	-1.054	-.817	-.584
.075	.143	-.146	-.610	-1.249	-.805	-.047	-.279	-.688	-.743	-.563
.150	.044	-.163	-.384	-.951	-.767	-.185	-.185	-.329	-.671	-.564
.250	-.029	-.177	-.333	-.699	-.727	-.075	-.098	-.122	-.596	-.563
.350	-.094	-.198	-.247	-.426	-.696					
.450	-.137	-.194	-.208	-.289	-.654					
.550	-.138	-.165	-.165	-.195	-.599					
.650	-.083	-.091	-.087	-.107	-.542	-.044	-.011	-.053	-.276	-.467
.750	-.016	-.001	-.011	-.027	-.470	-.004	.035	-.008	-.205	-.440
.850	.044	.059	.053	.039	-.385	.041	.085	.035	-.142	-.404
.900	.075	.089	.089	.072	-.335	.068	.102	.061	-.102	-.380
	Right side									
.025	-1.012	-.023	.346	.502	.591	-1.050	-.146	.177	.320	.391
.075	-.465	-.097	.159	.323	.450	-.561	-.257	-.057	.094	.196
.150	-.325	-.107	.073	.211	.327	-.206	-.187	-.208	-.115	-.013
.250	-.308	-.154	-.015	.102	.209					
.350	-.245	-.181	-.085	.008	.101					
.450	-.201	-.179	-.128	-.059	.014					
.550	-.151	-.146	-.120	-.087	-.046					
.650	-.076	-.103	-.071	-.062	-.065	-.028	-.006	-.038	-.080	-.158
.750	-.001	.025	-.014	-.020	-.075	.003	.027	-.016	-.078	-.210
.850	.084	.094	.068	.054	-.049	.071	.098	.057	-.015	-.186
.900	.119	.119	.095	.076	-.065	.091	.117	.073	.000	-.200

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL TAIL - Continued

(j) $\alpha = 8°$; M = 0.92

$\frac{x}{C_v}$	Cp for —					Cp for —				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.468	.919	.605	.414	-.064	.406	.687	.439	-.007	-.617
.025	-.196	-.134	.109	-.208	-.419	.316	-.069	-.985	-1.325	-1.189
.075	-.019	-.132	-.102	-.294	-.500	.159	-.112	-.445	-.865	-1.107
.150	.012	-.163	-.272	-.426	-.573	.079	-.117	-.341	-.675	-.988
.250	.010	-.137	-.205	-.382	-.543	.013	-.136	-.288	-.539	-.839
.350	-.024	-.136	-.207	-.325	-.396	-.060	-.180	-.304	-.394	-.627
.450	-.064	-.172	-.230	-.281	-.360	-.109	-.219	-.284	-.330	-.515
.550	-.081	-.180	-.204	-.214	-.353	-.105	-.171	-.182	-.255	-.457
.650	-.054	-.108	-.110	-.130	-.286	-.078	-.112	-.113	-.166	-.413
.750	-.025	-.062	-.053	-.050	-.218	-.025	-.030	-.033	-.075	-.346
.850	.031	.003	.018	.038	-.132	.052	.063	.050	.021	-.231
.900	.066	.045	.055	.086	-.066	.095	.105	.097	.072	-.149
Right side										
.025	.178	-.051	-.136	.145	.230	-.843	-.037	.360	.534	.653
.075	-.116	-.151	-.020	.160	.364	-.392	-.086	.176	.349	.497
.150	-.243	-.142	.025	.139	.398	-.306	-.096	.087	.237	.373
.250	-.193	-.128	.019	.119	.343	-.256	-.120	.022	.150	.263
.350	-.207	-.134	-.019	.093	.238	-.262	-.150	-.029	.076	.181
.450	-.206	-.159	-.064	.051	.147	-.244	-.180	-.090	-.004	.083
.550	-.177	-.162	-.077	.024	.088	-.168	-.163	-.103	-.036	.025
.650	-.098	-.104	-.053	.026	.057	-.095	-.103	-.073	-.036	-.002
.750	-.045	-.061	-.024	.028	.033	-.011	-.005	-.011	.003	-.005
.850	.035	.020	.039	.080	.057	.074	.081	.067	.068	.012
.900	.064	.046	.066	.091	.057	.106	.102	.085	.082	-.001
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.308	.607	.327	-.104	-.609	.312	.530	.199	-.203	-.603
.025	.286	-.124	-1.076	-1.280	-.802	.126	-.243	-1.045	-.784	-.541
.075	.141	-.150	-.635	-1.109	-.764	-.049	-.262	-.953	-.728	-.539
.150	.034	-.170	-.404	-.982	-.721	-.260	-.273	-.372	-.668	-.543
.250	-.041	-.193	-.371	-.747	-.891	-.076	-.065	-.101	-.602	-.541
.350	-.114	-.235	-.328	-.525	-.661					
.450	-.159	-.219	-.201	-.357	-.627					
.550	-.159	-.185	-.171	-.234	-.584					
.650	-.097	-.096	-.093	-.120	-.538	-.052	-.015	-.056	-.293	-.471
.750	-.021	-.005	-.012	-.024	-.483	-.006	.033	-.010	-.217	-.447
.850	.042	.055	.052	.039	-.411	.042	.074	.037	-.147	-.413
.900	.074	.093	.091	.074	-.376	.066	.101	.060	-.115	-.392
Right side										
.025	-1.040	-.062	.329	.500	.597	-1.085	-.185	.166	.321	.406
.075	-.432	-.126	.145	.324	.458	-.655	-.296	-.068	.091	.214
.150	-.333	-.130	.062	.210	.338	-.252	-.245	-.276	-.128	-.009
.250	-.338	-.176	-.032	.097	.215					
.350	-.289	-.219	-.113	.002	.105					
.450	-.201	-.206	-.163	-.067	.015					
.550	-.156	-.164	-.143	-.102	-.052					
.650	-.077	-.113	-.085	-.073	-.075	-.041	-.014	-.043	-.083	-.167
.750	-.003	.016	-.021	-.027	-.088	-.002	.027	-.018	-.086	-.217
.850	.084	.093	.067	.053	-.066	.068	.095	.055	-.020	-.206
.900	.116	.122	.097	.076	-.081	.089	.114	.072	-.007	-.210

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(k) $\alpha = 15.1°$; M = 0.60

	C_p for –					C_p for –				
$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.389	.843	.692	.631	.331	-.30?	.763	-.161	-1.557	-2.147
.025	-.507	-.260	.165	-.025	-.431	.37?	.022	-.627	-2.092	-2.154
.075	-.195	-.216	-.051	-.166	-.440	.23?	-.033	-.393	-.585	-2.343
.150	-.077	-.205	-.218	-.263	-.445	.154	-.046	-.280	-.485	-.595
.250	-.010	-.166	-.202	-.205	-.315	.084	-.062	-.234	-.407	-.512
.350	-.027	-.131	-.207	-.189	-.246	.024	-.099	-.218	-.367	-.493
.450	-.050	-.157	-.216	-.194	-.209	-.00?	-.111	-.202	-.342	-.452
.550	-.066	-.145	-.195	-.161	-.158	-.006	-.085	-.166	-.309	-.385
.650	-.066	-.115	-.152	-.119	-.098	.001	-.062	-.127	-.286	-.336
.750	-.059	-.083	-.106	-.085	-.038	.024	-.017	-.067	-.224	-.271
.850	-.034	-.042	-.046	-.038	.020	.06?	.036	-.007	-.131	-.214
.900	-.010	-.014	-.017	-.011	.038	.08?	.064	.029	-.082	-.161
Right side										
.025	.257	-.131	-.416	-.038	.165	-.79?	-.049	.366	.466	.507
.075	-.073	-.223	-.218	-.001	.195	-.41?	-.079	.208	.337	.458
.150	-.211	-.198	-.085	.045	.214	-.30?	-.074	.128	.246	.361
.250	-.209	-.161	-.017	.073	.158	-.24?	-.081	.080	.168	.278
.350	-.211	-.136	-.023	.047	.149	-.22?	-.099	.034	.112	.211
.450	-.223	-.159	-.049	.017	.158	-.20?	-.113	-.001	.059	.144
.550	-.197	-.145	-.051	-.001	.130	-.17?	-.104	-.010	.040	.110
.650	-.144	-.108	-.040	-.001	.119	-.12?	-.072	-.001	.036	.096
.750	-.100	-.076	-.028	.003	.103	-.06?	-.021	.027	.052	.096
.850	-.029	-.023	-.001	.031	.119	.001	.038	.073	.082	.107
.900	-.008	-.005	.009	.031	.105	.031	.050	.077	.087	.096
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	-.322	.493	-.264	-1.094	-1.140	-.02?	.426	-.310	-.809	-.678
.025	.350	-.021	-.856	-1.136	-1.098	.16?	-.108	-.640	-.879	-.659
.075	.209	-.053	-.601	-1.159	-1.193	-.00?	-.138	-.427	-.555	-.659
.150	.105	-.067	-.333	-1.191	-1.438	-.08?	-.120	-.221	-.497	-.639
.250	.036	-.079	-.223	-.582	-1.232	-.08?	-.074	-.147	-.460	-.602
.350	-.010	-.090	-.191	-.150	-.717					
.450	-.043	-.099	-.168	-.170	-.334					
.550	-.059	-.081	-.145	-.166	-.228					
.650	-.036	-.051	-.088	-.117	-.156	-.05?	-.014	-.074	-.286	-.588
.750	-.003	-.003	-.035	-.062	-.098	-.02?	.016	-.028	-.212	-.505
.850	.027	.041	.020	-.013	-.036	.00?	.048	.004	-.150	-.398
.900	.047	.061	.043	.019	-.013	.03?	.064	.032	-.110	-.322
Right side										
.025	-.835	-.062	.350	.448	.475	-.59?	-.145	.183	.246	.278
.075	-.641	-.095	.190	.316	.419	-.46?	-.163	-.028	.052	.130
.150	-.325	-.081	.105	.212	.324	-.33?	-.124	-.099	-.062	.010
.250	-.225	-.090	.025	.121	.218					
.350	-.202	-.104	-.019	.050	.140					
.450	-.170	-.113	-.042	-.001	.080					
.550	-.135	-.085	-.046	-.020	.043					
.650	-.087	-.046	-.028	-.015	.029	-.03?	-.012	-.058	-.089	-.121
.750	-.034	-.003	-.001	-.001	.027	-.02?	.011	-.028	-.092	-.117
.850	.036	.068	.055	.036	.047	.02?	.064	.025	-.043	-.084
.900	.068	.091	.077	.052	.045	.047	.077	.043	-.020	-.073

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(*l*) α = 15.1°; M = 0.80

x/cv	Cp for -					Cp for -				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
	z/bv = 0.11					z/bv = 0.38				
Left side										
.000	.564	.901	.802	.733	.399	.104	.826	.233	-.728	-1.492
.025	-.663	-.243	.210	.088	-.252	.374	.030	-.555	-1.504	-1.737
.075	-.195	-.218	-.014	-.094	-.343	.233	-.041	-.390	-.688	-1.840
.150	-.069	-.206	-.224	-.281	-.487	.164	-.052	-.295	-.541	-.768
.250	-.007	-.153	-.218	-.244	-.386	.096	-.066	-.247	-.449	-.551
.350	-.016	-.128	-.215	-.220	-.296	.034	-.096	-.232	-.402	-.528
.450	-.046	-.163	-.235	-.220	-.260	-.006	-.130	-.218	-.361	-.548
.550	-.057	-.153	-.209	-.178	-.207	-.006	-.095	-.175	-.327	-.445
.650	-.053	-.116	-.154	-.128	-.132	-.002	-.064	-.124	-.286	-.381
.750	-.045	-.081	-.100	-.083	-.065	.031	-.014	-.060	-.204	-.292
.850	-.021	-.032	-.035	-.025	.014	.078	.047	.011	-.095	-.215
.900	.007	.003	-.002	.008	.055	.101	.082	.052	-.034	-.155
Right side										
.025	.324	-.098	-.447	-.072	.138	-.729	-.052	.349	.464	.545
.075	-.022	-.227	-.223	-.009	.163	-.438	-.089	.207	.343	.462
.150	-.219	-.194	-.072	.048	.197	-.320	-.083	.140	.252	.376
.250	-.226	-.150	-.005	.083	.180	-.254	-.089	.088	.184	.295
.350	-.218	-.124	-.014	.066	.117	-.238	-.107	.041	.126	.221
.450	-.233	-.157	-.038	.028	.112	-.219	-.131	.002	.074	.154
.550	-.207	-.145	-.045	.006	.101	-.182	-.119	-.011	.048	.115
.650	-.144	-.099	-.031	.009	.091	-.127	-.078	.005	.046	.097
.750	-.089	-.070	-.014	.012	.083	-.051	-.022	.038	.066	.098
.850	-.019	-.008	.020	.041	.092	.019	.045	.088	.098	.111
.900	.005	.012	.031	.048	.084	.055	.059	.098	.109	.104
	z/bv = 0.66					z/bv = 0.93				
Left side										
.000	-.100	.535	-.029	-.793	-1.015	.004	.451	-.147	-.538	-.724
.025	.363	-.014	-.828	-1.166	-.965	.181	-.139	-.788	-.601	-.724
.075	.216	-.047	-.654	-1.178	-1.038	.002	-.160	-.716	-.571	-.727
.150	.119	-.070	-.378	-1.144	-1.169	-.103	-.136	-.240	-.527	-.716
.250	.049	-.083	-.227	-.767	-1.071	-.083	-.067	-.100	-.475	-.668
.350	-.009	-.099	-.194	-.138	-.833					
.450	-.045	-.105	-.172	-.155	-.605					
.550	-.059	-.087	-.135	-.154	-.436					
.650	-.031	-.046	-.071	-.095	-.275	-.042	.007	-.045	-.306	-.524
.750	.013	.023	-.014	-.037	-.164	-.006	.042	-.002	-.214	-.472
.850	.055	.065	.046	.021	-.065	.035	.079	.040	-.138	-.396
.900	.073	.090	.072	.045	-.028	.064	.102	.066	-.088	-.339
Right side										
.025	-.848	-.040	.375	.465	.505	-.896	-.142	.201	.281	.300
.075	-.690	-.078	.206	.329	.427	-.371	-.177	-.017	.080	.155
.150	-.382	-.067	.118	.237	.339	-.318	-.130	-.100	-.049	.006
.250	-.226	-.089	.051	.137	.235					
.350	-.198	-.107	-.014	.057	.147					
.450	-.163	-.101	-.040	.009	.081					
.550	-.124	-.080	-.043	-.015	.038					
.650	-.068	-.037	-.022	-.009	.018	-.007	.019	-.011	-.088	-.137
.750	-.009	.018	.015	.008	.015	.008	.042	-.002	-.083	-.143
.850	.070	.093	.078	.058	.038	.064	.103	.058	-.026	-.106
.900	.101	.119	.106	.074	.038	.086	.116	.071	-.002	-.108

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Concluded

(m) $\alpha = 15.1°$; M = 0.85

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v=0.11$					$z/b_v=0.38$				
Left side										
.000	.599	.924	.838	.749	.401	.192	.843	.344	-.566	-1.224
.025	-.714	-.223	.214	.113	-.180	.355	.018	-.513	-1.356	-1.597
.075	-.216	-.219	-.005	-.076	-.276	.216	-.052	-.407	-.722	-1.630
.150	-.082	-.210	-.231	-.287	-.461	.153	-.063	-.315	-.593	-.826
.250	-.016	-.159	-.239	-.278	-.444	.085	-.078	-.266	-.492	-.590
.350	-.032	-.133	-.233	-.244	-.331	.024	-.110	-.256	-.431	-.511
.450	-.059	-.176	-.263	-.248	-.283	-.019	-.146	-.239	-.387	-.649
.550	-.069	-.159	-.226	-.198	-.243	-.016	-.110	-.190	-.344	-.476
.650	-.062	-.123	-.167	-.139	-.173	-.011	-.076	-.139	-.301	-.425
.750	-.051	-.083	-.110	-.097	-.095	.019	-.021	-.062	-.212	-.321
.850	-.025	-.028	-.039	-.028	-.002	.068	.048	.008	-.100	-.230
.900	-.001	.009	-.005	.002	.044	.097	.085	.056	-.038	-.167
Right side										
.025	.338	-.096	-.498	-.089	.116	-.722	-.058	.333	.455	.546
.075	-.014	-.241	-.233	-.025	.140	-.471	-.096	.199	.337	.461
.150	-.234	-.190	-.078	.041	.182	-.346	-.088	.136	.252	.372
.250	-.249	-.136	-.009	.080	.170	-.279	-.090	.091	.183	.291
.350	-.237	-.125	-.021	.057	.110	-.257	-.116	.038	.123	.219
.450	-.252	-.159	-.051	.019	.085	-.237	-.139	-.008	.068	.149
.550	-.224	-.149	-.058	-.001	.078	-.194	-.123	-.021	.041	.108
.650	-.154	-.099	-.039	-.001	.064	-.133	-.085	-.006	.036	.084
.750	-.104	-.068	-.025	-.002	.058	-.059	-.026	.036	.061	.088
.850	-.026	-.008	.014	.035	.077	.024	.048	.086	.094	.096
.900	.002	.016	.031	.038	.068	.054	.061	.102	.101	.090

$\frac{x}{c_v}$	$z/b_v=0.66$					$z/b_v=0.93$				
Left side										
.000	-.061	.546	.032	-.686	-1.031	.032	.454	-.115	-.544	-.729
.025	.355	-.025	-.882	-1.209	-.962	.170	-.146	-.360	-.636	-.731
.075	.212	-.058	-.706	-1.221	-.989	-.014	-.184	-.770	-.599	-.708
.150	.108	-.076	-.395	-1.172	-1.016	-.137	-.147	-.290	-.558	-.695
.250	.037	-.093	-.250	-.794	-.966	-.110	-.075	-.096	-.505	-.668
.350	-.022	-.109	-.213	-.172	-.831					
.450	-.065	-.116	-.182	-.195	-.675					
.550	-.081	-.093	-.146	-.185	-.517					
.650	-.045	-.049	-.082	-.116	-.354	-.049	.014	-.051	-.318	-.519
.750	.005	.028	-.015	-.051	-.239	-.011	.045	.001	-.231	-.465
.850	.048	.073	.048	.013	-.127	.035	.092	.051	-.149	-.398
.900	.070	.096	.078	.039	-.080	.065	.112	.078	-.103	-.349
Right side										
.025	-.952	-.036	.375	.460	.506	-.859	-.139	.199	.285	.308
.075	-.794	-.080	.204	.325	.428	-.509	-.196	-.033	.078	.157
.150	-.411	-.070	.122	.232	.337	-.349	-.145	-.125	-.066	.004
.250	-.237	-.093	.044	.136	.230					
.350	-.214	-.110	-.021	.049	.137					
.450	-.181	-.113	-.053	-.004	.070					
.550	-.137	-.088	-.056	-.034	.021					
.650	-.080	-.042	-.029	-.027	.001	-.028	.014	-.015	-.109	-.168
.750	-.014	.022	.015	-.002	-.006	.002	.051	.001	-.097	-.174
.850	.067	.105	.088	.051	.019	.063	.115	.065	-.037	-.140
.900	.100	.130	.106	.064	.018	.080	.129	.085	-.014	-.141

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$

(a) $\alpha = -8°$; $M = 0.60$

$\dfrac{x}{c_v}$	C_p for—					C_p for—				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.683	.825	.474	-.359	-1.024	.407	.876	-.499	-1.132	-.785
.025	.264	-.145	-.671	-1.258	-1.373	.365	-.115	-.720	-1.139	-1.107
.075	.103	-.170	-.457	-.766	-1.316	.135	-.186	-.552	-1.049	-1.143
.150	-.017	-.214	-.411	-.587	-1.118	.024	-.207	-.439	-.748	-.939
.250	-.056	-.195	-.317	-.398	-.833	-.070	-.234	-.404	-.500	-.909
.350	-.082	-.184	-.275	-.300	-.587	-.139	-.267	-.404	-.410	-.766
.450	-.132	-.207	-.271	-.265	-.417	-.181	-.280	-.386	-.352	-.596
.550	-.128	-.184	-.229	-.215	-.302	-.176	-.248	-.319	-.297	-.454
.650	-.093	-.122	-.162	-.148	-.215	-.144	-.193	-.241	-.251	-.352
.750	-.047	-.076	-.095	-.097	-.159	-.095	-.120	-.151	-.196	-.288
.850	-.001	-.012	-.019	-.031	-.104	-.008	-.028	-.049	-.129	-.244
.900	.029	.029	.027	.008	-.067	.020	.011	.001	-.086	-.208
Right side										
.025	-.579	-.101	.303	.590	.818	-.985	-.129	.386	.641	.811
.075	-.499	-.198	.077	.305	.526	-.568	-.202	.133	.349	.542
.150	-.416	-.212	-.017	.153	.339	-.464	-.218	.013	.181	.362
.250	-.333	-.202	-.056	.073	.225	-.402	-.239	-.075	.064	.215
.350	-.296	-.195	-.095	-.001	.119	-.393	-.262	-.137	-.031	.105
.450	-.287	-.216	-.135	-.067	.034	-.370	-.278	-.188	-.097	.015
.550	-.236	-.193	-.128	-.086	-.001	-.314	-.255	-.171	-.123	-.033
.650	-.160	-.131	-.089	-.063	-.010	-.234	-.191	-.142	-.118	-.056
.750	-.091	-.074	-.040	-.035	-.005	-.137	-.111	-.079	-.081	-.056
.850	-.006	-.001	.017	.013	.015	-.031	-.017	.001	-.028	-.040
.900	.022	.034	.041	.031	.018	.006	.011	.024	-.019	-.067
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.379	.876	.462	-.024	-.428	.428	.743	.252	.199	-.274
.025	.393	-.177	-1.135	-1.086	-.833	.296	-.521	-1.669	-.440	-.403
.075	.149	-.239	-.681	-1.065	-.856	-.006	-.588	-1.347	-.438	-.405
.150	-.010	-.292	-.600	-.861	-.881	-.250	-.627	-1.324	-.426	-.412
.250	-.125	-.340	-.554	-.532	-.838	-.324	-.544	-1.093	-.433	-.426
.350	-.208	-.377	-.547	-.392	-.686					
.450	-.264	-.390	-.515	-.371	-.502					
.550	-.261	-.349	-.432	-.375	-.488					
.650	-.208	-.248	-.314	-.398	-.516	-.245	-.280	-.294	-.470	-.481
.750	-.137	-.161	-.204	-.435	-.532	-.146	-.152	-.185	-.470	-.477
.850	-.056	-.076	-.098	-.424	-.513	-.056	-.040	-.091	-.438	-.435
.900	-.031	-.030	-.047	-.405	-.511	-.015	.006	-.042	-.410	-.417
Right side										
.025	-1.026	-.166	.418	.657	.802	-1.586	-.487	.326	.553	.799
.075	-.711	-.260	.133	.351	.542	-1.328	-.615	-.031	.188	.461
.150	-.579	-.283	.001	.179	.362	-1.365	-.601	-.231	-.054	.197
.250	-.570	-.345	-.123	.018	.197					
.350	-.568	-.381	-.206	-.100	.064					
.450	-.515	-.381	-.252	-.180	-.042					
.550	-.427	-.342	-.248	-.208	-.104					
.650	-.317	-.255	-.197	-.196	-.129	-.277	-.239	-.190	-.238	-.182
.750	-.199	-.168	-.123	-.173	-.146	-.183	-.143	-.128	-.221	-.215
.850	-.072	-.044	-.024	-.123	-.136	-.077	-.017	-.015	-.171	-.182
.900	-.026	-.005	.004	-.116	-.173	-.031	.032	.017	-.187	-.201

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(b) $\alpha = -9°$; M = 0.80

$\frac{x}{c_v}$	C_p for- $z/b_v = 0.11$					C_p for- $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.840	.891	.697	.282	-.215	.678	.919	-.759	.374	-.085
.025	.276	-.110	-.619	-1.099	-.963	.362	-.087	-1.129	-1.612	-.638
.075	.117	-.149	-.448	-.955	-.860	.141	-.182	-.469	-1.433	-.644
.150	-.010	-.221	-.428	-.830	-.790	.021	-.216	-.423	-1.258	-.666
.250	-.063	-.218	-.314	-.372	-.674	-.083	-.262	-.380	-.317	-.695
.350	-.100	-.213	-.267	-.247	-.600	-.189	-.328	-.389	-.293	-.664
.450	-.177	-.270	-.282	-.236	-.534	-.273	-.389	-.380	-.302	-.588
.550	-.200	-.257	-.236	-.203	-.461	-.291	-.362	-.313	-.264	-.490
.650	-.155	-.164	-.156	-.143	-.389	-.253	-.279	-.239	-.233	-.417
.750	-.091	-.096	-.084	-.097	-.313	-.148	-.142	-.158	-.193	-.343
.850	-.015	-.009	-.006	-.034	-.232	-.048	-.029	-.056	-.140	-.288
.900	.027	.036	.043	.002	-.190	.004	.026	.000	-.097	-.255
Right side										
.025	-.492	-.053	.319	.599	.830	-1.076	-.109	.378	.640	.819
.075	-.495	-.190	.092	.320	.542	-.530	-.201	.132	.359	.560
.150	-.423	-.219	-.009	.175	.360	-.427	-.227	.013	.200	.380
.250	-.326	-.221	-.066	.083	.239	-.371	-.267	-.083	.070	.230
.350	-.281	-.230	-.115	-.005	.125	-.366	-.328	-.179	-.047	.105
.450	-.279	-.273	-.187	-.094	.022	-.362	-.383	-.262	-.149	-.008
.550	-.235	-.260	-.202	-.138	-.041	-.310	-.359	-.279	-.207	-.085
.650	-.149	-.168	-.145	-.112	-.050	-.230	-.248	-.234	-.196	-.117
.750	-.074	-.087	-.087	-.086	-.057	-.138	-.127	-.132	-.143	-.111
.850	.010	.008	.002	-.021	-.044	-.039	-.010	-.026	-.073	-.099
.900	.047	.040	.029	.002	-.054	.005	.029	-.001	-.059	-.126

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.698	.936	.773	.403	.008	.722	.825	.684	.392	-.097
.025	.378	-.124	-1.214	-1.634	-.727	.239	-.418	-1.201	-.377	-.398
.075	.151	-.210	-.808	-1.400	-.741	-.054	-.570	-.521	-.360	-.385
.150	-.015	-.293	-.509	-.901	-.761	-.346	-.799	-.480	-.346	-.381
.250	-.160	-.389	-.445	-.591	-.706	-.363	-.716	-.474	-.351	-.386
.350	-.294	-.515	-.445	-.342	-.658					
.450	-.432	-.612	-.438	-.297	-.553					
.550	-.482	-.540	-.389	-.322	-.455					
.650	-.385	-.253	-.331	-.363	-.458	-.351	-.294	-.426	-.388	-.429
.750	-.193	-.173	-.271	-.398	-.472	-.242	-.156	-.414	-.394	-.423
.850	-.128	-.060	-.224	-.420	-.466	-.175	-.043	-.374	-.377	-.401
.900	-.106	-.014	-.193	-.430	-.472	-.154	.009	-.339	-.369	-.389
Right side										
.025	-1.196	-.118	.407	.667	.825	-1.100	-.360	.255	.591	.837
.075	-.799	-.237	.131	.375	.571	-.440	-.685	-.099	.218	.504
.150	-.472	-.285	-.007	.198	.391	-.418	-.765	-.331	-.050	.218
.250	-.433	-.405	-.168	.025	.216					
.350	-.432	-.537	-.299	-.132	.059					
.450	-.409	-.589	-.418	-.268	-.082					
.550	-.359	-.512	-.443	-.342	-.175					
.650	-.308	-.305	-.360	-.322	-.210	-.397	-.259	-.322	-.346	-.261
.750	-.274	-.170	-.179	-.252	-.200	-.395	-.149	-.216	-.270	-.245
.850	-.230	-.029	-.090	-.144	-.160	-.368	-.026	-.121	-.181	-.178
.900	-.198	.025	-.069	-.167	-.187	-.337	.025	-.116	-.183	-.189

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(c) $\alpha = -9°$; $M = 0.85$

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.11$					C_p for — $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.877	.914	.755	.391	-.063	.734	.938	.818	.492	.074
.025	.286	-.092	-.556	-1.022	-.771	.363	-.075	-1.022	-1.415	-.598
.075	.129	-.137	-.503	-.885	-.706	.144	-.173	-.788	-1.284	-.608
.150	-.008	-.219	-.437	-.833	-.681	.027	-.209	-.427	-1.193	-.628
.250	-.060	-.217	-.315	-.688	-.618	-.080	-.251	-.368	-.816	-.650
.350	-.106	-.213	-.269	-.251	-.586	-.187	-.325	-.383	-.178	-.636
.450	-.197	-.296	-.289	-.199	-.563	-.297	-.421	-.399	-.212	-.588
.550	-.253	-.321	-.249	-.179	-.506	-.354	-.464	-.328	-.237	-.517
.650	-.214	-.213	-.161	-.142	-.448	-.371	-.407	-.246	-.231	-.454
.750	-.151	-.112	-.088	-.103	-.383	-.276	-.125	-.162	-.205	-.396
.850	-.040	-.008	-.008	-.045	-.307	-.067	-.016	-.065	-.159	-.353
.900	.012	.046	.042	-.005	-.274	-.011	.041	-.007	-.117	-.316
Right side										
.025	-.446	-.025	.329	.600	.827	-.997	-.076	.379	.637	.816
.075	-.578	-.172	.102	.322	.543	-.793	-.184	.135	.360	.557
.150	-.468	-.211	-.004	.170	.357	-.426	-.213	.018	.205	.380
.250	-.333	-.216	-.065	.081	.236	-.354	-.249	-.087	.074	.232
.350	-.284	-.230	-.121	-.010	.123	-.357	-.324	-.185	-.050	.097
.450	-.287	-.290	-.208	-.116	.004	-.371	-.417	-.296	-.175	-.030
.550	-.243	-.313	-.256	-.175	-.067	-.316	-.474	-.360	-.255	-.124
.650	-.150	-.203	-.206	-.156	-.090	-.233	-.307	-.359	-.274	-.170
.750	-.078	-.099	-.136	-.133	-.113	-.141	-.122	-.198	-.211	-.173
.850	.009	.011	-.025	-.053	-.094	-.053	-.001	-.059	-.104	-.146
.900	.047	.053	.016	-.021	-.113	-.010	.045	-.015	-.086	-.174

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.752	.958	.829	.509	.122	.770	.861	.722	.419	.026
.025	.390	-.098	-1.101	-1.469	-.694	.250	-.345	-1.007	-.879	-.416
.075	.167	-.184	-.905	-1.310	-.698	-.018	-.488	-.447	-.363	-.398
.150	.004	-.264	-.596	-1.179	-.706	-.330	-.733	-.406	-.349	-.390
.250	-.143	-.354	-.368	-.481	-.660	-.320	-.730	-.402	-.350	-.397
.350	-.286	-.482	-.399	-.347	-.590					
.450	-.431	-.622	-.439	-.344	-.530					
.550	-.561	-.724	-.385	-.320	-.468					
.650	-.633	-.639	-.326	-.349	-.473	-.858	-.418	-.407	-.380	-.426
.750	-.511	-.085	-.291	-.386	-.483	-.370	-.281	-.405	-.395	-.423
.850	-.097	-.043	-.259	-.409	-.473	-.211	-.123	-.375	-.374	-.404
.900	-.108	-.005	-.241	-.426	-.484	-.181	-.059	-.356	-.363	-.396
Right side										
.025	-1.078	-.076	.411	.667	.826	-.991	-.267	.280	.614	.843
.075	-.900	-.204	.142	.380	.573	-.386	-.580	-.074	.248	.510
.150	-.651	-.249	.008	.209	.393	-.363	-.692	-.312	-.031	.220
.250	-.337	-.361	-.151	.035	.214					
.350	-.394	-.495	-.298	-.127	.049					
.450	-.404	-.629	-.439	-.285	-.106					
.550	-.354	-.716	-.566	-.395	-.230					
.650	-.304	-.529	-.597	-.468	-.311	-.370	-.375	-.609	-.660	-.480
.750	-.291	-.119	-.420	-.429	-.310	-.378	-.273	-.463	-.393	-.290
.850	-.273	-.019	-.074	-.135	-.181	-.361	-.139	-.269	-.212	-.187
.900	-.250	.024	-.082	-.165	-.204	-.340	-.070	-.205	-.185	-.187

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0^0$ - Continued

(d) $\alpha = -9^0$; $M = 0.90$

$\frac{x}{C_v}$	C_p for–					C_p for–				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.915	.937	.809	.497	.057	.794	.954	.865	.597	.167
.025	.303	-.071	-.482	-.921	-.821	.374	-.054	-.898	-1.209	-.613
.075	.147	-.119	-.483	-.785	-.736	.162	-.161	-.745	-1.117	-.618
.150	.008	-.211	-.449	-.770	-.702	.046	-.200	-.563	-1.055	-.647
.250	-.051	-.212	-.490	-.776	-.636	-.056	-.234	-.521	-.979	-.664
.350	-.089	-.199	-.264	-.520	-.578	-.169	-.310	-.358	-.674	-.640
.450	-.196	-.299	-.291	-.203	-.543	-.283	-.395	-.324	-.101	-.581
.550	-.308	-.377	-.252	-.120	-.486	-.375	-.493	-.345	-.147	-.502
.650	-.264	-.304	-.167	-.108	-.425	-.429	-.547	-.259	-.201	-.441
.750	-.312	-.173	-.098	-.099	-.363	-.449	-.393	-.172	-.214	-.409
.850	-.134	-.009	-.012	-.056	-.295	-.386	.029	-.081	-.195	-.390
.900	-.024	.048	.037	-.022	-.266	-.075	.068	-.020	-.153	-.348
Right side										
.025	-.365	-.002	.347	.615	.841	-.859	-.064	.383	.641	.829
.075	-.518	-.156	.113	.340	.557	-.754	-.177	.148	.372	.575
.150	-.459	-.207	.003	.187	.374	-.607	-.203	.030	.217	.398
.250	-.503	-.214	-.056	.095	.252	-.492	-.234	-.066	.083	.249
.350	-.288	-.215	-.110	-.001	.134	-.334	-.307	-.171	-.039	.114
.450	-.268	-.277	-.198	-.110	.009	-.284	-.390	-.286	-.170	-.024
.550	-.235	-.401	-.325	-.206	-.079	-.319	-.508	-.395	-.263	-.129
.650	-.148	-.292	-.267	-.210	-.117	-.239	-.571	-.433	-.341	-.203
.750	-.082	-.139	-.283	-.260	-.174	-.152	-.157	-.452	-.376	-.252
.850	.007	.012	-.087	-.149	-.168	-.066	.020	-.191	-.293	-.256
.900	.050	.056	-.018	-.087	-.189	-.024	.059	-.072	-.228	-.289
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.800	.977	.878	.602	.206	.813	.893	.784	.561	.124
.025	.408	-.067	-.979	-1.263	-.731	.304	-.254	-.872	-.856	-.402
.075	.190	-.160	-.829	-1.139	-.737	.036	-.419	-.401	-.347	-.382
.150	.036	-.235	-.763	-1.062	-.725	-.281	-.664	-.356	-.340	-.378
.250	-.106	-.322	-.652	-.848	-.640	-.307	-.687	-.351	-.340	-.378
.350	-.253	-.440	-.151	-.405	-.502					
.450	-.400	-.571	-.336	-.349	-.479					
.550	-.521	-.702	-.394	-.344	-.440					
.650	-.638	-.705	-.322	-.360	-.447	-.881	-.447	-.378	-.367	-.402
.750	-.677	-.511	-.325	-.386	-.455	-.846	-.431	-.386	-.372	-.401
.850	-.626	-.010	-.318	-.405	-.447	-.554	-.330	-.368	-.360	-.386
.900	-.515	.022	-.306	-.422	-.455	-.412	-.260	-.355	-.353	-.381
Right side										
.025	-.942	-.053	.424	.672	.845	-.859	-.195	.318	.633	.863
.075	-.807	-.181	.167	.395	.592	-.351	-.496	-.035	.279	.534
.150	-.725	-.218	.033	.230	.414	-.322	-.625	-.278	.007	.251
.250	-.654	-.323	-.121	.061	.244					
.350	-.136	-.453	-.270	-.102	.072					
.450	-.300	-.588	-.401	-.260	-.089					
.550	-.339	-.687	-.552	-.405	-.220					
.650	-.289	-.689	-.639	-.482	-.320	-.335	-.411	-.779	-.741	-.564
.750	-.313	-.353	-.695	-.562	-.426	-.347	-.372	-.779	-.783	-.031
.850	-.319	.017	-.620	-.572	-.443	-.339	-.284	-.508	-.610	-.459
.900	-.311	.037	-.340	-.564	-.436	-.326	-.226	-.421	-.429	-.279

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(c) $\alpha = -9°$; M = 0.92

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	C_p for — $z/b_v = 0.11$					C_p for — $z/b_v = 0.38$				
	Left side									
.000	.931	.950	.829	.522	.100	.809	.963	.883	.621	.204
.025	.300	-.061	-.452	-.896	-.809	.367	-.041	-.849	-1.170	-.604
.075	.147	-.113	-.464	-.758	-.718	.159	-.160	-.715	-1.084	-.611
.150	.006	-.209	-.454	-.744	-.686	.043	-.202	-.579	-1.021	-.628
.250	-.051	-.222	-.520	-.756	-.624	-.055	-.227	-.503	-.956	-.648
.350	-.095	-.199	-.328	-.576	-.567	-.167	-.304	-.489	-.833	-.633
.450	-.199	-.301	-.311	-.234	-.534	-.279	-.374	-.334	-.122	-.586
.550	-.308	-.373	-.234	-.104	-.491	-.381	-.485	-.262	-.100	-.511
.650	-.271	-.357	-.164	-.086	-.441	-.456	-.567	-.283	-.168	-.454
.750	-.347	-.350	-.106	-.086	-.389	-.458	-.534	-.196	-.197	-.418
.850	-.228	-.043	-.020	-.056	-.330	-.482	-.057	-.101	-.186	-.396
.900	-.074	.037	.030	-.020	-.291	-.372	.081	-.040	-.154	-.359
	Right side									
.025	-.348	.003	.351	.623	.851	-.824	-.067	.389	.646	.839
.075	-.517	-.158	.121	.348	.566	-.751	-.186	.155	.377	.586
.150	-.461	-.213	.013	.196	.385	-.619	-.211	.038	.227	.408
.250	-.543	-.230	-.048	.103	.267	-.554	-.235	-.057	.095	.260
.350	-.345	-.214	-.103	.008	.147	-.462	-.307	-.162	-.026	.125
.450	-.300	-.303	-.189	-.102	.019	-.281	-.393	-.270	-.158	-.014
.550	-.220	-.411	-.336	-.196	-.077	-.263	-.492	-.391	-.249	-.120
.650	-.154	-.340	-.271	-.198	-.112	-.256	-.587	-.460	-.329	-.195
.750	-.095	-.291	-.340	-.275	-.188	-.178	-.494	-.445	-.371	-.256
.850	-.006	-.012	-.175	-.174	-.188	-.098	-.017	-.433	-.333	-.262
.900	.030	.047	-.058	-.108	-.250	-.051	.053	-.209	-.306	-.324
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.818	.986	.899	.625	.240	.825	.908	.803	.575	.118
.025	.405	-.053	-.924	-1.226	-.707	.312	-.218	-.826	-.921	-.404
.075	.193	-.150	-.781	-1.102	-.707	.047	-.382	-.391	-.331	-.385
.150	.038	-.227	-.734	-1.042	-.709	-.267	-.635	-.344	-.329	-.378
.250	-.102	-.313	-.658	-.925	-.647	-.309	-.664	-.338	-.330	-.378
.350	-.247	-.424	-.416	-.426	-.538					
.450	-.392	-.551	-.228	-.331	-.504					
.550	-.514	-.684	-.330	-.329	-.441					
.650	-.638	-.744	-.336	-.347	-.439	-.895	-.480	-.371	-.357	-.401
.750	-.698	-.691	-.351	-.371	-.446	-.840	-.461	-.380	-.358	-.402
.850	-.660	-.230	-.351	-.387	-.439	-.594	-.422	-.370	-.350	-.393
.900	-.664	-.051	-.344	-.413	-.450	-.494	-.371	-.360	-.355	-.384
	Right side									
.025	-.905	-.051	.433	.679	.854	-.835	-.174	.338	.643	.875
.075	-.789	-.182	.179	.404	.606	-.364	-.469	-.007	.292	.553
.150	-.718	-.219	.049	.244	.429	-.331	-.613	-.258	.021	.274
.250	-.686	-.316	-.102	.075	.257					
.350	-.417	-.439	-.254	-.086	.089					
.450	-.208	-.572	-.380	-.244	-.069					
.550	-.308	-.687	-.527	-.385	-.204					
.650	-.313	-.734	-.622	-.469	-.305	-.355	-.476	-.770	-.724	-.548
.750	-.347	-.628	-.698	-.547	-.412	-.383	-.445	-.801	-.784	-.611
.850	-.356	-.098	-.665	-.559	-.435	-.355	-.379	-.572	-.658	-.565
.900	-.364	.014	-.642	-.587	-.466	-.348	-.324	-.475	-.522	-.463

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(f) $\alpha = 0°$; $M = 0.60$

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.615	.640	.311	-.441	-1.142	.245	.714	.297	-.420	-.973
.025	.266	-.103	-.549	-1.088	-1.760	.336	-.085	-.649	-1.034	-.898
.075	.132	-.110	-.368	-.647	-1.344	.159	-.123	-.449	-.907	-.875
.150	.041	-.137	-.309	-.459	-.827	.066	-.132	-.336	-.666	-.820
.250	.021	-.098	-.214	-.286	-.398	-.002	-.146	-.289	-.400	-.767
.350	-.024	-.110	-.187	-.222	-.307	-.058	-.166	-.284	-.304	-.669
.450	-.056	-.128	-.178	-.200	-.273	-.083	-.182	-.262	-.259	-.551
.550	-.058	-.107	-.137	-.150	-.262	-.083	-.157	-.212	-.213	-.444
.650	-.033	-.069	-.087	-.107	-.187	-.063	-.119	-.151	-.172	-.364
.750	-.011	-.031	-.033	-.068	-.122	-.029	-.060	-.078	-.118	-.278
.850	.030	.028	.023	-.016	-.049	.030	.019	.005	-.047	-.198
.900	.048	.048	.051	.012	-.002	.057	.055	.051	-.011	-.135
	Right side									
.025	-.462	-.028	.315	.584	.820	-.643	-.067	.381	.600	.749
.075	-.389	-.107	.127	.339	.551	-.430	-.121	.166	.371	.547
.150	-.299	-.119	.055	.216	.396	-.335	-.132	.071	.232	.405
.250	-.222	-.105	.023	.143	.296	-.281	-.144	.005	.134	.287
.350	-.194	-.121	-.022	.073	.200	-.267	-.164	-.047	.064	.205
.450	-.183	-.132	-.063	.014	.120	-.253	-.177	-.078	.005	.125
.550	-.138	-.134	-.063	-.007	.089	-.210	-.159	-.083	-.022	.076
.650	-.088	-.076	-.033	-.002	.080	-.145	-.119	-.065	-.016	.058
.750	-.031	-.028	-.011	.016	.073	-.072	-.065	-.020	.000	.049
.850	.035	.030	.035	.048	.089	.016	.024	.041	.039	.058
.900	.050	.053	.051	.053	.089	.035	.044	.053	.046	.040
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.198	.721	.207	-.295	-.533	.393	.775	.098	-.227	-.727
.025	.379	-.110	-.877	-.952	-.618	.484	-.252	-1.185	-.852	-.538
.075	.202	-.137	-.535	-.904	-.633	.205	-.315	-.988	-.343	-.362
.150	.089	-.157	-.425	-.813	-.651	-.013	-.333	-1.013	-.336	-.358
.250	.007	-.175	-.377	-.588	-.615	-.092	-.304	-.818	-.341	-.367
.350	-.063	-.211	-.368	-.372	-.560					
.450	-.106	-.227	-.341	-.272	-.480					
.550	-.120	-.216	-.280	-.261	-.407					
.650	-.092	-.153	-.198	-.263	-.382	-.108	-.159	-.171	-.382	-.402
.750	-.047	-.080	-.106	-.272	-.362	-.056	-.074	-.085	-.377	-.371
.850	.010	-.006	-.020	-.247	-.327	.012	.014	.005	-.347	-.313
.900	.025	.026	.014	-.229	-.304	.035	.053	.041	-.316	-.294
	Right side									
.025	-.823	-.055	.426	.630	.736	-1.183	-.166	.526	.723	.909
.075	-.519	-.125	.207	.403	.569	-1.018	-.290	.200	.403	.640
.150	-.385	-.134	.105	.266	.433	-1.011	-.288	.017	.199	.418
.250	-.367	-.177	.019	.150	.316					
.350	-.369	-.207	-.054	.062	.213					
.450	-.326	-.227	-.097	-.009	.131					
.550	-.276	-.209	-.115	-.050	.067					
.650	-.192	-.162	-.087	-.057	.029	-.165	-.155	-.097	-.104	-.024
.750	-.106	-.078	-.045	-.043	.002	-.083	-.078	-.078	-.122	-.073
.850	-.002	.010	.028	-.011	-.011	.019	.024	.026	-.088	-.062
.900	.035	.042	.035	-.020	-.047	.055	.069	.055	-.113	-.093

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(g) $\alpha = 0°$; $M = 0.80$

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.741	.668	.514	.181	-.215	.480	.775	.549	.043	-.396
.025	.270	-.068	-.534	-1.143	-.895	.337	-.068	-.914	-1.583	-.620
.075	.140	-.095	-.363	-.846	-.802	.158	-.122	-.431	-1.120	-.619
.150	.045	-.131	-.315	-.486	-.731	.069	-.137	-.326	-.725	-.626
.250	.022	-.101	-.211	-.282	-.622	.000	-.155	-.273	-.383	-.628
.350	-.023	-.121	-.179	-.227	-.529	-.067	-.190	-.270	-.283	-.594
.450	-.070	-.146	-.182	-.199	-.465	-.123	-.217	-.265	-.236	-.534
.550	-.079	-.128	-.142	-.149	-.392	-.124	-.195	-.211	-.190	-.465
.650	-.055	-.083	-.083	-.106	-.314	-.100	-.143	-.154	-.158	-.408
.750	-.024	-.033	-.026	-.058	-.244	-.049	-.060	-.075	-.117	-.339
.850	.030	.030	.032	-.010	-.151	.024	.029	.014	-.047	-.282
.900	.056	.068	.069	.022	-.103	.065	.074	.063	-.003	-.242
	Right side									
.025	-.397	.003	.324	.591	.815	-.916	-.045	.393	.611	.749
.075	-.384	-.095	.142	.357	.556	-.419	-.109	.175	.378	.545
.150	-.299	-.110	.068	.233	.397	-.323	-.125	.080	.249	.400
.250	-.219	-.101	.030	.161	.295	-.272	-.146	.009	.146	.280
.350	-.195	-.124	-.023	.077	.189	-.265	-.176	-.050	.066	.192
.450	-.189	-.145	-.071	.010	.101	-.263	-.204	-.103	-.005	.095
.550	-.147	-.131	-.081	-.017	.052	-.224	-.188	-.122	-.049	.034
.650	-.083	-.077	-.053	-.014	.032	-.153	-.137	-.092	-.050	.010
.750	-.023	-.029	-.023	-.005	.020	-.061	-.048	-.035	-.028	-.006
.850	.042	.041	.039	.034	.025	.030	.047	.038	.016	-.015
.900	.074	.066	.057	.046	.017	.063	.065	.056	.018	-.048

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side									
.000	.467	.783	.508	.093	-.285	.666	.837	.561	.178	-.376
.025	.380	-.084	-1.093	-.822	-.619	.430	-.277	-.864	-.828	-.432
.075	.204	-.122	-.486	-.802	-.634	.161	-.401	-.409	-.297	-.382
.150	.087	-.160	-.363	-.719	-.652	-.068	-.413	-.406	-.291	-.371
.250	.000	-.204	-.300	-.545	-.644	-.142	-.357	-.401	-.298	-.383
.350	-.091	-.265	-.303	-.359	-.616					
.450	-.166	-.302	-.308	-.267	-.520					
.550	-.194	-.274	-.280	-.252	-.440					
.650	-.151	-.185	-.214	-.261	-.423	-.168	-.160	-.309	-.344	-.420
.750	-.076	-.077	-.143	-.271	-.417	-.107	-.047	-.283	-.351	-.405
.850	-.008	.009	-.071	-.264	-.396	-.042	.057	-.234	-.336	-.367
.900	.021	.053	-.030	-.250	-.380	-.021	.103	-.198	-.321	-.344
	Right side									
.025	-1.046	-.030	.433	.641	.752	-.935	-.139	.490	.757	.941
.075	-.459	-.109	.211	.410	.576	-.552	-.374	.155	.429	.655
.150	-.340	-.130	.112	.285	.442	-.542	-.344	-.026	.195	.407
.250	-.333	-.195	.006	.164	.316					
.350	-.352	-.252	-.084	.052	.195					
.450	-.340	-.287	-.155	-.041	.090					
.550	-.287	-.267	-.187	-.102	.005					
.650	-.204	-.208	-.151	-.114	-.043	-.262	-.160	-.172	-.170	-.109
.750	-.123	-.057	-.087	-.100	-.072	-.225	-.060	-.130	-.174	-.147
.850	-.018	.044	.002	-.056	-.072	-.160	.057	-.041	-.120	-.121
.900	.027	.075	.030	-.067	-.112	-.123	.097	-.039	-.141	-.154

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = 0° - Continued

(h) α = 0°; M = 0.85

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.11$					C_p for — $z/b_v = 0.38$				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
Left side										
.000	.774	.676	.563	.301	-.058	.536	.784	.607	.199	-.226
.025	.272	-.069	-.527	-1.057	-.834	.336	-.078	-1.028	-1.470	-.558
.075	.146	-.100	-.375	-.827	-.719	.158	-.136	-.410	-1.308	-.558
.150	.046	-.146	-.336	-.655	-.653	.070	-.152	-.330	-.864	-.567
.250	.019	-.117	-.222	-.293	-.570	-.007	-.173	-.278	-.254	-.572
.350	-.029	-.137	-.192	-.213	-.509	-.077	-.214	-.275	-.287	-.556
.450	-.089	-.173	-.194	-.200	-.478	-.145	-.258	-.276	-.256	-.515
.550	-.103	-.159	-.151	-.155	-.424	-.157	-.243	-.219	-.206	-.465
.650	-.083	-.110	-.095	-.115	-.362	-.143	-.182	-.164	-.169	-.416
.750	-.053	-.053	-.034	-.075	-.304	-.082	-.080	-.085	-.127	-.359
.850	.013	.024	.030	-.019	-.219	.002	.024	.006	-.060	-.314
.900	.046	.068	.063	.016	-.171	.051	.078	.056	-.019	-.276
Right side										
.025	-.376	.006	.323	.587	.823	-.949	-.042	.371	.599	.759
.075	-.389	-.100	.139	.354	.562	-.419	-.120	.169	.372	.550
.150	-.314	-.119	.065	.229	.399	-.328	-.135	.076	.240	.402
.250	-.226	-.115	.023	.158	.295	-.272	-.157	-.001	.141	.283
.350	-.205	-.143	-.034	.071	.189	-.267	-.196	-.069	.061	.190
.450	-.196	-.170	-.091	-.006	.091	-.269	-.234	-.134	-.029	.086
.550	-.155	-.163	-.108	-.040	.036	-.229	-.231	-.156	-.078	.019
.650	-.093	-.106	-.078	-.037	.008	-.158	-.176	-.131	-.087	-.020
.750	-.029	-.049	-.047	-.036	-.012	-.069	-.066	-.068	-.064	-.040
.850	.040	.035	.025	.017	-.012	.022	.041	.022	-.009	-.047
.900	.073	.063	.048	.025	-.026	.059	.068	.040	-.010	-.087

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.66$					C_p for — $z/b_v = 0.93$				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
Left side										
.000	.522	.793	.573	.176	-.178	.717	.865	.628	.251	-.255
.025	.384	-.089	-1.155	-.967	-.595	.438	-.254	-.810	-.859	-.421
.075	.207	-.130	-.416	-.902	-.601	.158	-.458	-.352	-.303	-.380
.150	.090	-.170	-.346	-.761	-.618	-.083	-.471	-.349	-.296	-.363
.250	-.002	-.226	-.289	-.534	-.612	-.155	-.358	-.348	-.300	-.372
.350	-.110	-.307	-.293	-.342	-.609					
.450	-.208	-.367	-.308	-.266	-.536					
.550	-.252	-.362	-.283	-.257	-.443					
.650	-.211	-.209	-.221	-.269	-.423	-.219	-.153	-.308	-.340	-.413
.750	-.123	-.090	-.164	-.281	-.424	-.147	-.041	-.302	-.350	-.407
.850	-.039	.019	-.104	-.279	-.411	-.083	.066	-.263	-.342	-.378
.900	-.009	.059	-.065	-.267	-.400	-.066	.100	-.228	-.328	-.361
Right side										
.025	-1.114	-.029	.424	.639	.764	-.879	-.119	.488	.773	.953
.075	-.356	-.112	.207	.409	.584	-.409	-.427	.145	.436	.663
.150	-.322	-.135	.106	.287	.447	-.399	-.380	-.042	.190	.411
.250	-.304	-.211	-.002	.165	.320					
.350	-.322	-.286	-.109	.044	.194					
.450	-.320	-.350	-.201	-.065	.077					
.550	-.280	-.345	-.252	-.148	-.024					
.650	-.216	-.268	-.213	-.166	-.082	-.300	-.197	-.216	-.217	-.150
.750	-.153	-.076	-.125	-.145	-.116	-.283	-.072	-.158	-.212	-.190
.850	-.060	.044	-.021	-.082	-.102	-.238	.053	-.074	-.146	-.144
.900	-.022	.081	.005	-.091	-.139	-.201	.099	-.068	-.163	-.176

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(i) $\alpha = 0°$; M = 0.90

$\frac{x}{c_v}$	C_p for - $z/b_v = 0.11$					C_p for - $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.804	.687	.615	.401	.075	.587	.802	.673	.329	-.119
.025	.283	-.054	-.478	-.953	-.819	.337	-.063	-.950	-1.281	-.542
.075	.152	-.090	-.381	-.751	-.688	.162	-.130	-.632	-1.166	-.540
.150	.045	-.147	-.368	-.660	-.637	.069	-.147	-.347	-1.055	-.550
.250	.016	-.119	-.238	-.469	-.558	-.008	-.166	-.272	-.484	-.556
.350	-.035	-.141	-.202	-.244	-.500	-.085	-.218	-.273	-.127	-.539
.450	-.105	-.190	-.203	-.164	-.477	-.176	-.289	-.281	-.208	-.507
.550	-.133	-.202	-.163	-.133	-.433	-.208	-.317	-.230	-.195	-.464
.650	-.125	-.161	-.103	-.111	-.381	-.219	-.279	-.170	-.167	-.422
.750	-.095	-.078	-.041	-.069	-.327	-.147	-.101	-.096	-.124	-.371
.850	-.019	.019	.030	-.013	-.257	-.031	.034	-.006	-.063	-.327
.900	.027	.070	.066	.019	-.217	.028	.090	.048	-.016	-.289
Right side										
.025	-.346	.019	.327	.595	.834	-.916	-.037	.368	.606	.768
.075	-.417	-.094	.142	.364	.571	-.614	-.114	.166	.380	.560
.150	-.351	-.121	.060	.238	.410	-.348	-.129	.073	.250	.410
.250	-.245	-.115	.021	.167	.305	-.265	-.155	-.002	.147	.292
.350	-.214	-.145	-.041	.074	.195	-.264	-.205	-.077	.060	.190
.450	-.203	-.185	-.107	-.008	.087	-.269	-.269	-.155	-.033	.084
.550	-.163	-.194	-.140	-.057	.026	-.233	-.290	-.201	-.096	.006
.650	-.099	-.142	-.123	-.063	-.018	-.160	-.242	-.209	-.128	-.045
.750	-.034	-.067	-.093	-.072	-.042	-.075	-.089	-.140	-.121	-.078
.850	.040	.035	-.007	-.016	-.046	.012	.055	-.025	-.057	-.087
.900	.070	.071	.024	.000	-.070	.046	.083	.013	-.037	-.125

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.571	.815	.637	.288	-.081	.763	.897	.691	.336	-.127
.025	.390	-.065	-1.060	-1.347	-.585	.455	-.182	-.825	-.837	-.454
.075	.215	-.110	-.860	-1.194	-.590	.171	-.430	-.285	-.297	-.386
.150	.097	-.149	-.287	-.734	-.605	-.083	-.486	-.284	-.287	-.370
.250	.000	-.207	-.227	-.506	-.602	-.139	-.453	-.287	-.291	-.372
.350	-.118	-.323	-.270	-.369	-.609					
.450	-.238	-.426	-.303	-.287	-.543					
.550	-.350	-.473	-.273	-.248	-.446					
.650	-.334	-.441	-.225	-.244	-.423	-.581	-.246	-.291	-.319	-.407
.750	-.254	-.066	-.179	-.270	-.430	-.191	-.133	-.295	-.329	-.409
.850	-.067	.038	-.134	-.281	-.422	-.094	-.011	-.274	-.322	-.390
.900	-.031	.081	-.101	-.276	-.418	-.087	.045	-.253	-.315	-.375
Right side										
.025	-1.047	-.011	.425	.645	.782	-.754	-.066	.500	.789	.970
.075	-.799	-.095	.213	.429	.594	-.311	-.453	.151	.449	.673
.150	-.252	-.118	.115	.300	.460	-.307	-.411	-.040	.209	.415
.250	-.252	-.202	.001	.178	.330					
.350	-.293	-.310	-.111	.051	.198					
.450	-.300	-.379	-.227	-.069	.066					
.550	-.269	-.426	-.327	-.171	-.048					
.650	-.218	-.382	-.338	-.247	-.131	-.293	-.247	-.562	-.454	-.240
.750	-.171	-.034	-.317	-.276	-.191	-.293	-.158	-.256	-.362	-.293
.850	-.099	.063	-.054	-.135	-.148	-.264	-.043	-.104	-.148	-.182
.900	-.071	.101	-.018	-.113	-.174	-.235	.026	-.087	-.153	-.211

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(j) $\alpha = 0°$; M = 0.92

$z/b_v = 0.11$ and $z/b_v = 0.38$ — Left side

$\dfrac{x}{c_v}$	C_p for — $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for — $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.819	.688	.629	.431	.104	.612	.806	.689	.364	-.090
.025	.277	-.060	-.452	-.920	-.819	.334	-.073	-.879	-1.220	-.563
.075	.151	-.094	-.374	-.729	-.682	.156	-.141	-.686	-1.114	-.559
.150	.039	-.158	-.375	-.653	-.627	.067	-.151	-.394	-1.018	-.559
.250	.009	-.126	-.256	-.536	-.548	-.010	-.168	-.271	-.689	-.552
.350	-.039	-.146	-.202	-.308	-.494	-.091	-.224	-.266	-.138	-.534
.450	-.115	-.201	-.205	-.176	-.469	-.189	-.297	-.275	-.164	-.497
.550	-.151	-.227	-.165	-.127	-.426	-.224	-.341	-.236	-.180	-.454
.650	-.153	-.196	-.107	-.105	-.375	-.264	-.352	-.170	-.163	-.418
.750	-.135	-.102	-.039	-.075	-.319	-.224	-.137	-.095	-.131	-.363
.850	-.043	.014	.029	-.019	-.254	-.066	.034	-.002	-.073	-.321
.900	.011	.065	.066	.016	-.209	.007	.089	.051	-.030	-.275

Right side

$\dfrac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	-.334	.023	.331	.603	.838	-.885	-.036	.368	.611	.772
.075	-.412	-.094	.147	.372	.579	-.637	-.121	.171	.385	.565
.150	-.370	-.126	.064	.242	.417	-.396	-.134	.077	.256	.416
.250	-.268	-.117	.023	.171	.313	-.270	-.155	.002	.151	.296
.350	-.228	-.153	-.041	.079	.205	-.269	-.205	-.070	.064	.200
.450	-.216	-.192	-.104	-.011	.093	-.273	-.277	-.153	-.034	.085
.550	-.172	-.215	-.148	-.061	.032	-.250	-.320	-.206	-.098	.007
.650	-.107	-.168	-.137	-.072	-.008	-.172	-.293	-.231	-.142	-.048
.750	-.042	-.085	-.119	-.093	-.042	-.084	-.120	-.189	-.154	-.082
.850	.033	.027	-.021	-.038	-.049	.007	.053	-.054	-.100	-.096
.900	.062	.067	.012	-.019	-.069	.041	.080	-.001	-.072	-.132

$z/b_v = 0.66$ and $z/b_v = 0.93$ — Left side

$\dfrac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.592	.821	.656	.323	-.056	.780	.912	.708	.368	-.090
.025	.389	-.071	-.984	-1.219	-.592	.464	-.168	-.796	-.779	-.455
.075	.217	-.114	-.830	-1.079	-.598	.172	-.427	-.271	-.298	-.377
.150	.096	-.151	-.428	-.850	-.612	-.086	-.497	-.271	-.289	-.361
.250	.002	-.205	-.172	-.517	-.603	-.133	-.472	-.272	-.289	-.363
.350	-.119	-.322	-.240	-.371	-.603					
.450	-.242	-.438	-.300	-.299	-.539					
.550	-.359	-.511	-.275	-.244	-.435					
.650	-.414	-.529	-.223	-.248	-.411	-.642	-.277	-.277	-.316	-.402
.750	-.380	-.094	-.178	-.273	-.419	-.520	-.199	-.284	-.327	-.409
.850	-.144	.040	-.136	-.290	-.417	-.135	-.081	-.266	-.318	-.387
.900	-.056	.080	-.104	-.289	-.413	-.103	-.023	-.246	-.311	-.375

Right side

$\dfrac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	-1.003	-.006	.433	.655	.784	-.719	-.044	.508	.797	.975
.075	-.792	-.098	.221	.435	.599	-.301	-.445	.162	.455	.675
.150	-.301	-.117	.121	.307	.464	-.294	-.410	-.038	.213	.418
.250	-.222	-.198	.010	.180	.337					
.350	-.277	-.312	-.103	.053	.207					
.450	-.313	-.401	-.226	-.072	.069					
.550	-.275	-.458	-.328	-.175	-.046					
.650	-.220	-.426	-.378	-.254	-.133	-.264	-.254	-.617	-.495	-.281
.750	-.180	-.088	-.410	-.319	-.213	-.291	-.187	-.478	-.512	-.343
.850	-.112	.072	-.109	-.265	-.165	-.268	-.085	-.133	-.270	-.198
.900	-.084	.098	-.027	-.186	-.185	-.241	-.024	-.099	-.168	-.224

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(k) $\alpha = 9°$; M = 0.60

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.472	.822	.580	.036	-1.140	.018	.646	-.096	-1.036	-1.402
.025	-.085	-.108	-.048	-.446	-.925	.366	-.029	-.768	-1.294	-1.161
.075	.041	-.101	-.163	-.395	-.631	.210	-.050	-.402	-1.402	-1.127
.150	.055	-.115	-.200	-.342	-.480	.137	-.060	-.265	-.381	-1.108
.250	.055	-.085	-.138	-.234	-.331	.080	-.060	-.214	-.243	-.796
.350	.027	-.076	-.133	-.192	-.276	.038	-.078	-.194	-.243	-.480
.450	.004	-.090	-.133	-.167	-.246	.011	-.092	-.170	-.206	-.347
.550	-.001	-.078	-.110	-.130	-.216	.013	-.060	-.124	-.151	-.296
.650	.016	-.039	-.059	-.091	-.175	.018	-.039	-.082	-.107	-.267
.750	.029	-.016	-.025	-.036	-.136	.036	.003	-.041	-.059	-.228
.850	.057	.024	.029	.024	-.058	.077	.066	.040	.004	-.157
.900	.075	.047	.054	.066	-.007	.096	.091	.077	.043	-.111
	Right side									
.025	.027	-.064	-.020	.211	.717	-.601	-.002	.422	.577	.660
.075	-.163	-.122	.047	.204	.541	-.356	-.036	.239	.405	.536
.150	-.186	-.104	.068	.211	.424	-.255	-.046	.156	.292	.417
.250	-.140	-.081	.068	.213	.343	-.195	-.050	.098	.211	.321
.350	-.138	-.076	.040	.170	.254	-.175	-.057	.063	.160	.254
.450	-.134	-.085	.015	.123	.183	-.159	-.062	.029	.103	.174
.550	-.101	-.074	.008	.100	.149	-.120	-.053	.019	.077	.132
.650	-.058	-.041	.017	.091	.128	-.072	-.027	.029	.068	.103
.750	-.021	-.006	.040	.087	.110	-.017	.024	.052	.073	.080
.850	.038	.038	.070	.112	.114	.052	.080	.093	.100	.077
.900	.061	.049	.089	.114	.112	.075	.084	.100	.091	.050
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	-.040	.604	-.117	-.683	-.918	.337	.799	-.214	-.404	-.425
.025	.412	-.036	-.872	-.883	-.615	.662	.059	-.925	-.446	-.397
.075	.270	-.041	-.552	-.849	-.608	.449	-.034	-.849	-.436	-.411
.150	.183	-.041	-.316	-.805	-.590	.233	-.076	-.714	-.413	-.407
.250	.130	-.039	-.235	-.674	-.560	.137	-.048	-.471	-.381	-.402
.350	.084	-.062	-.210	-.457	-.540					
.450	.041	-.078	-.194	-.273	-.503					
.550	.006	-.081	-.157	-.185	-.455					
.650	.013	-.046	-.099	-.137	-.423	-.019	-.060	-.096	-.234	-.345
.750	.036	-.002	-.032	-.100	-.381	.009	-.013	-.032	-.206	-.331
.850	.055	.045	.024	-.068	-.349	.036	.033	.031	-.178	-.308
.900	.075	.073	.059	-.042	-.333	.057	.059	.061	-.162	-.299
	Right side									
.025	-.792	.024	.457	.607	.655	-.962	.121	.718	.872	.963
.075	-.386	-.018	.290	.451	.568	-.716	.010	.478	.651	.802
.150	-.257	-.016	.212	.349	.474	-.689	-.025	.281	.437	.580
.250	-.218	-.027	.156	.280	.392					
.350	-.207	-.048	.093	.195	.291					
.450	-.186	-.067	.056	.133	.210					
.550	-.154	-.062	.017	.077	.130					
.650	-.092	-.036	.015	.047	.068	-.085	-.064	.003	.015	-.005
.750	-.028	-.009	.040	.045	.025	-.019	-.011	.003	-.031	-.099
.850	.048	.084	.091	.064	-.005	.059	.063	.066	-.022	-.136
.900	.077	.093	.098	.068	-.046	.087	.096	.087	-.038	-.182

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(i) $\alpha = 9°$; $M = 0.80$

$\frac{x}{c_v}$	\multicolumn{5}{c}{C_p for -}	\multicolumn{5}{c}{C_p for -}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	\multicolumn{5}{c}{$z/b_v = 0.11$}	\multicolumn{5}{c}{$z/b_v = 0.38$}
	Left side
.000	.479	.890	.596	.235	-.496	.276	.692	.241	-.402	-.970
.025	-.107	-.108	.022	-.365	-.731	.379	-.032	-.837	-1.294	-.976
.075	.046	-.101	-.141	-.389	-.630	.224	-.058	-.424	-1.279	-.991
.150	.065	-.116	-.219	-.377	-.493	.148	-.054	-.267	-.647	-.947
.250	.064	-.086	-.152	-.253	-.377	.099	-.058	-.215	-.211	-.807
.350	.040	-.073	-.151	-.202	-.304	.050	-.081	-.207	-.236	-.568
.450	.011	-.095	-.155	-.181	-.268	.017	-.099	-.192	-.210	-.399
.550	.002	-.078	-.118	-.138	-.224	.017	-.072	-.143	-.156	-.318
.650	.012	-.041	-.063	-.087	-.190	.020	-.046	-.089	-.109	-.288
.750	.029	-.019	-.027	-.047	-.161	.040	.004	-.027	-.060	-.251
.850	.069	.032	.033	.017	-.096	.090	.074	.050	.003	-.193
.900	.096	.065	.065	.060	-.044	.117	.111	.093	.046	-.146
	Right side
.025	.102	-.041	-.044	.220	.572	-.674	.027	.433	.585	.690
.075	-.132	-.116	.050	.212	.514	-.350	-.026	.247	.411	.559
.150	-.184	-.093	.079	.186	.443	-.242	-.032	.166	.305	.446
.250	-.137	-.069	.079	.197	.370	-.190	-.038	.115	.233	.353
.350	-.145	-.073	.048	.158	.280	-.180	-.054	.079	.170	.279
.450	-.148	-.092	.015	.117	.201	-.169	-.067	.042	.109	.201
.550	-.107	-.075	.008	.095	.158	-.131	-.060	.025	.077	.148
.650	-.058	-.046	.022	.083	.123	-.072	-.029	.030	.063	.110
.750	-.014	-.014	.033	.080	.110	-.001	.020	.059	.062	.079
.850	.050	.045	.080	.105	.107	.075	.096	.105	.092	.064
.900	.079	.058	.094	.106	.093	.108	.108	.105	.083	.027
	\multicolumn{5}{c}{$z/b_v = 0.66$}	\multicolumn{5}{c}{$z/b_v = 0.93$}
	Left side
.000	.219	.655	.167	-.303	-.811	.562	.878	.187	-.110	-.638
.025	.423	-.023	-.808	-.714	-.589	.677	.020	-.849	-.432	-.393
.075	.288	-.032	-.592	-.685	-.575	.470	-.063	-.741	-.395	-.402
.150	.213	-.032	-.339	-.632	-.556	.247	-.115	-.710	-.357	-.400
.250	.169	-.031	-.241	-.575	-.522	.140	-.078	-.600	-.342	-.394
.350	.104	-.064	-.238	-.512	-.504					
.450	.041	-.096	-.229	-.382	-.486					
.550	.002	-.101	-.186	-.260	-.449					
.650	.005	-.063	-.109	-.181	-.425	-.037	-.069	-.076	-.224	-.361
.750	.032	-.002	-.027	-.139	-.399	-.005	-.003	-.007	-.205	-.349
.850	.065	.062	.042	-.110	-.373	.040	.067	.054	-.191	-.332
.900	.090	.088	.079	-.096	-.362	.069	.100	.086	-.179	-.321
	Right side
.025	-.799	.064	.475	.626	.706	-.892	.149	.742	.904	1.030
.075	-.425	.007	.310	.477	.615	-.644	.004	.496	.677	.851
.150	-.231	.009	.241	.379	.519	-.661	-.022	.296	.451	.624
.250	-.215	-.014	.178	.309	.441					
.350	-.222	-.044	.115	.217	.332					
.450	-.204	-.075	.053	.137	.239					
.550	-.158	-.081	.010	.074	.148					
.650	-.085	-.054	.015	.036	.075	.015	-.031	-.010	-.020	-.008
.750	-.011	-.011	.034	.017	.012	.017	.003	.001	-.081	-.125
.850	.082	.105	.089	.032	-.023	.087	.087	.062	-.081	-.184
.900	.117	.117	.106	.029	-.079	.120	.119	.083	-.104	-.253

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(m) $\alpha = 9°$; $M = 0.85$

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.478	.901	.287	.274	-.384	.327	.707	-.233	-.260	-.947
.025	-.124	-.109	-.321	-.334	-.648	.370	-.041	-1.300	-1.308	-.938
.075	.036	-.105	-.377	-.388	-.611	.219	-.068	-1.183	-1.152	-.921
.150	.060	-.121	-.407	-.422	-.517	.149	-.064	-.719	-.766	-.912
.250	.063	-.094	-.271	-.281	-.382	.097	-.067	-.234	-.267	-.811
.350	.032	-.084	-.214	-.225	-.328	.048	-.089	-.264	-.277	-.612
.450	.000	-.104	-.200	-.213	-.291	.009	-.109	-.246	-.257	-.431
.550	-.008	-.087	-.164	-.173	-.238	.006	-.087	-.199	-.213	-.336
.650	-.001	-.048	-.119	-.131	-.204	.009	-.057	-.159	-.175	-.296
.750	.018	-.029	-.079	-.091	-.168	.029	-.008	-.109	-.122	-.259
.850	.060	.032	-.004	-.017	-.108	.080	.073	-.032	-.044	-.202
.900	.089	.063	.042	.031	-.053	.109	.112	.019	-.004	-.151
Right side										
.025	.122	-.042	.212	.209	.547	-.806	.035	.588	.594	.697
.075	-.136	-.118	.210	.212	.473	-.365	-.027	.410	.415	.564
.150	-.202	-.099	.183	.185	.433	-.253	-.035	.310	.305	.451
.250	-.154	-.072	.190	.193	.378	-.204	-.044	.236	.236	.358
.350	-.161	-.077	.157	.156	.290	-.192	-.058	.173	.170	.291
.450	-.168	-.101	.109	.105	.209	-.191	-.077	.110	.105	.212
.550	-.128	-.082	.088	.084	.165	-.151	-.072	.075	.068	.158
.650	-.071	-.051	.073	.068	.126	-.091	-.038	.056	.050	.115
.750	-.025	-.021	.076	.065	.108	-.011	.009	.052	.047	.085
.850	.042	.040	.096	.088	.103	.069	.093	.072	.064	.062
.900	.072	.063	.095	.088	.089	.103	.106	.065	.051	.025
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.270	.672	-.184	-.212	-.749	.624	.888	-.062	-.104	-.615
.025	.426	-.028	-.646	-.645	-.679	.682	-.015	-.499	-.523	-.413
.075	.287	-.035	-.616	-.627	-.625	.468	-.081	-.417	-.415	-.416
.150	.212	-.035	-.563	-.582	-.588	.239	-.139	-.367	-.371	-.411
.250	.166	-.038	-.482	-.494	-.554	.136	-.101	-.358	-.367	-.410
.350	.096	-.081	-.434	-.436	-.524					
.450	.032	-.116	-.401	-.399	-.497					
.550	-.018	-.124	-.355	-.361	-.460					
.650	-.019	-.081	-.304	-.315	-.430	-.065	-.084	-.327	-.342	-.374
.750	.012	-.015	-.249	-.271	-.410	-.029	-.014	-.297	-.323	-.362
.850	.052	.055	-.186	-.213	-.384	.022	.070	-.251	-.283	-.347
.900	.079	.093	-.153	-.198	-.373	.050	.103	-.226	-.260	-.336
Right side										
.025	-.850	.072	.632	.638	.722	-.855	.159	.915	.919	1.045
.075	-.453	.018	.485	.486	.627	-.640	-.007	.686	.686	.866
.150	-.236	.016	.387	.389	.532	-.631	-.027	.459	.457	.640
.250	-.221	-.018	.316	.312	.455					
.350	-.245	-.052	.222	.221	.351					
.450	-.238	-.089	.140	.128	.251					
.550	-.185	-.098	.066	.062	.160					
.650	-.105	-.068	.024	.018	.086	-.055	-.048	-.039	-.044	-.013
.750	-.022	-.022	-.008	-.019	.020	.000	.003	-.116	-.129	-.111
.850	.070	.103	.008	-.007	-.026	.068	.095	-.116	-.135	-.165
.900	.107	.120	-.002	-.024	-.080	.099	.129	-.146	-.178	-.244

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(n) $\alpha = 9°$; M = 0.90

$\frac{x}{c_v}$	\multicolumn{5}{c}{C_p for — $z/b_v = 0.11$}	\multicolumn{5}{c}{C_p for — $z/b_v = 0.38$}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side
.000	.496	.926	.618	.340	-.417	.369	.712	.408	-.078	-.723
.025	-.115	-.117	.080	-.262	-.518	.380	-.053	-.978	-1.355	-.980
.075	.044	-.116	-.111	-.333	-.554	.228	-.080	-.426	-1.042	-.935
.150	.070	-.135	-.238	-.424	-.585	.157	-.077	-.268	-.489	-.888
.250	.072	-.107	-.160	-.285	-.382	.109	-.078	-.207	-.332	-.795
.350	.041	-.096	-.152	-.214	-.348	.056	-.105	-.204	-.272	-.632
.450	.012	-.125	-.166	-.201	-.329	.019	-.140	-.198	-.242	-.480
.550	.005	-.111	-.132	-.165	-.274	.012	-.120	-.158	-.205	-.384
.650	.012	-.072	-.081	-.123	-.235	.013	-.092	-.111	-.170	-.337
.750	.021	-.052	-.043	-.084	-.201	.034	-.036	-.048	-.119	-.293
.850	.066	.013	.019	-.013	-.130	.087	.054	.036	-.047	-.235
.900	.095	.046	.060	.032	-.091	.115	.099	.084	.005	-.188
Right side
.025	.148	-.048	-.059	.203	.484	-.839	.028	.431	.592	.710
.075	-.118	-.134	.042	.210	.486	-.348	-.037	.250	.414	.568
.150	-.188	-.111	.079	.189	.455	-.244	-.044	.171	.312	.458
.250	-.140	-.089	.082	.186	.391	-.188	-.054	.122	.241	.366
.350	-.146	-.089	.046	.155	.296	-.179	-.074	.080	.182	.299
.450	-.156	-.113	.007	.108	.210	-.178	-.099	.031	.114	.218
.550	-.112	-.097	.002	.088	.165	-.143	-.096	.008	.071	.159
.650	-.059	-.074	-.001	.072	.119	-.082	-.070	.007	.051	.119
.750	-.016	-.042	.020	.064	.096	.002	-.015	.030	.045	.080
.850	.058	.028	.063	.087	.089	.081	.081	.076	.060	.053
.900	.086	.048	.079	.084	.067	.109	.093	.084	.045	.010

$\frac{x}{c_v}$	\multicolumn{5}{c}{$z/b_v = 0.66$}	\multicolumn{5}{c}{$z/b_v = 0.93$}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side
.000	.307	.680	.318	-.123	-.677	.664	.913	.457	-.025	-.610
.025	.437	-.033	-.982	-.706	-.720	.687	-.019	-.711	-.758	-.452
.075	.305	-.038	-.563	-.657	-.653	.476	-.107	-.397	-.357	-.460
.150	.226	-.042	-.260	-.594	-.612	.243	-.174	-.358	-.328	-.449
.250	.179	-.056	-.187	-.482	-.586	.140	-.128	-.350	-.324	-.451
.350	.108	-.111	-.198	-.392	-.546					
.450	.035	-.158	-.218	-.327	-.526					
.550	-.015	-.174	-.210	-.272	-.495					
.650	-.018	-.131	-.153	-.249	-.471	-.065	-.123	-.240	-.352	-.417
.750	.010	-.045	-.087	-.244	-.448	-.030	-.029	-.196	-.353	-.404
.850	.056	.040	-.014	-.216	-.424	.017	.058	-.137	-.327	-.386
.900	.081	.076	.028	-.195	-.416	.042	.093	-.101	-.308	-.374
Right side
.025	-.840	.069	.483	.643	.736	-.794	.154	.732	.930	1.061
.075	-.437	.010	.321	.494	.638	-.575	-.040	.484	.697	.882
.150	-.217	.009	.253	.402	.540	-.560	-.050	.285	.470	.651
.250	-.204	-.026	.190	.329	.466					
.350	-.232	-.078	.118	.232	.360					
.450	-.234	-.124	.042	.142	.262					
.550	-.186	-.147	-.020	.062	.167					
.650	-.104	-.121	-.040	.011	.089	-.066	-.076	-.071	-.059	.010
.750	-.016	-.062	-.028	-.027	.016	-.009	-.025	-.095	-.134	-.097
.850	.083	.089	.036	-.013	-.022	.059	.084	-.047	-.134	-.132
.900	.118	.108	.056	-.032	-.078	.091	.115	-.041	-.186	-.205

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; i_t = 0° - Continued

(o) α = 9°; M = 0.92

$\frac{x}{c_v}$	Cp for - β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	Cp for - β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
	z/b_v = 0.11					z/b_v = 0.38				
	Left side									
.000	.491	.935	.621	.346	-.383	.416	.716	.435	-.022	-.673
.025	-.143	-.101	.086	-.248	-.481	.374	-.037	-.990	-1.340	-1.013
.075	.028	-.102	-.106	-.325	-.531	.220	-.069	-.428	-.964	-.959
.150	.059	-.125	-.247	-.437	-.583	.148	-.065	-.277	-.586	-.887
.250	.057	-.095	-.169	-.319	-.405	.100	-.069	-.211	-.374	-.789
.350	.032	-.084	-.162	-.239	-.342	.045	-.095	-.211	-.329	-.629
.450	-.006	-.117	-.176	-.231	-.336	-.001	-.136	-.204	-.271	-.487
.550	-.021	-.108	-.143	-.191	-.278	-.013	-.122	-.172	-.229	-.397
.650	-.016	-.071	-.091	-.148	-.253	-.022	-.096	-.124	-.194	-.358
.750	-.004	-.046	-.052	-.111	-.219	-.006	-.035	-.061	-.145	-.313
.850	.044	.020	.008	-.037	-.154	.053	.060	.028	-.069	-.253
.900	.078	.059	.047	.009	-.105	.090	.106	.074	-.021	-.204
	Right side									
.025	.163	-.031	-.077	.183	.464	-.872	.035	.420	.581	.713
.075	-.112	-.122	.029	.191	.481	-.355	-.027	.242	.400	.571
.150	-.212	-.096	.070	.176	.458	-.259	-.033	.161	.296	.458
.250	-.159	-.074	.071	.167	.396	-.195	-.040	.111	.228	.367
.350	-.166	-.074	.037	.138	.298	-.185	-.062	.073	.171	.298
.450	-.173	-.108	.002	.095	.213	-.189	-.091	.024	.101	.219
.550	-.133	-.089	-.007	.070	.168	-.164	-.102	-.003	.062	.163
.650	-.081	-.070	-.013	.055	.118	-.105	-.072	-.006	.037	.117
.750	-.038	-.041	.008	.047	.097	-.026	-.018	.015	.025	.080
.850	.036	.032	.052	.068	.083	.057	.082	.064	.039	.050
.900	.062	.056	.070	.062	.061	.092	.102	.073	.024	.005
	z/b_v = 0.66					z/b_v = 0.93				
	Left side									
.000	.353	.692	.347	-.088	-.649	.723	.922	.489	.007	-.593
.025	.430	-.017	-1.015	-.804	-.734	.684	.011	-.701	-.845	-.449
.075	.298	-.027	-.546	-.721	-.658	.463	-.105	-.371	-.383	-.471
.150	.223	-.030	-.261	-.643	-.616	.228	-.173	-.344	-.354	-.462
.250	.162	-.041	-.187	-.528	-.592	.123	-.110	-.338	-.351	-.465
.350	.092	-.102	-.198	-.401	-.556					
.450	.012	-.157	-.229	-.322	-.531					
.550	-.052	-.179	-.220	-.274	-.506					
.650	-.068	-.143	-.166	-.261	-.481	-.135	-.128	-.245	-.377	-.430
.750	-.042	-.041	-.102	-.258	-.465	-.105	-.027	-.214	-.386	-.417
.850	.007	.045	-.030	-.236	-.440	-.075	.067	-.165	-.365	-.401
.900	.033	.086	.010	-.221	-.433	-.062	.106	-.133	-.347	-.395
	Right side									
.025	-.956	.084	.480	.639	.742	-.650	.173	.727	.930	1.070
.075	-.397	.025	.313	.482	.641	-.380	-.032	.476	.690	.887
.150	-.192	.025	.252	.394	.547	-.363	-.046	.278	.457	.658
.250	-.172	-.014	.182	.316	.469					
.350	-.208	-.061	.106	.223	.364					
.450	-.229	-.118	.031	.126	.265					
.550	-.209	-.147	-.035	.043	.166					
.650	-.141	-.128	-.060	-.016	.090	-.197	-.078	-.097	-.079	.016
.750	-.068	-.071	-.051	-.049	.017	-.159	-.010	-.118	-.153	-.098
.850	.027	.097	.020	-.030	-.020	-.112	.089	-.069	-.145	-.123
.900	.065	.116	.039	-.057	-.075	-.073	.119	-.073	-.203	-.193

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued

(p) α = 15.1°; M = 0.60

$\frac{x}{c_v}$	C_p for $z/b_v = 0.11$ β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	C_p for $z/b_v = 0.38$ β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
Left side										
.000	.408	.851	.706	.595	.309	-.314	.796	-.232	-1.556	-2.058
.025	-.450	-.207	.174	-.030	-.449	.418	.038	-.643	-2.050	-2.060
.075	-.151	-.177	-.043	-.155	-.449	.275	-.003	-.373	-.526	-2.364
.150	-.035	-.167	-.195	-.241	-.447	.208	-.001	-.251	-.424	-.490
.250	.033	-.131	-.179	-.180	-.312	.156	-.003	-.200	-.348	-.529
.350	.021	-.089	-.177	-.157	-.252	.112	-.024	-.191	-.317	-.472
.450	.003	-.107	-.186	-.159	-.215	.082	-.045	-.170	-.285	-.421
.550	-.016	-.098	-.158	-.125	-.167	.077	-.024	-.133	-.250	-.376
.650	-.035	-.068	-.112	-.084	-.110	.070	-.008	-.094	-.230	-.337
.750	-.025	-.040	-.075	-.057	-.067	.077	.022	-.040	-.187	-.284
.850	-.002	-.003	-.027	-.016	-.007	.098	.066	.015	-.134	-.238
.900	.017	.017	.008	.014	.032	.110	.089	.057	-.084	-.193
Right side										
.025	.264	-.121	-.371	.016	.208	-.778	.024	.412	.517	.574
.075	-.053	-.191	-.172	.055	.235	-.382	-.017	.287	.401	.524
.150	-.186	-.161	-.040	.096	.251	-.258	-.013	.195	.319	.441
.250	-.179	-.117	.027	.128	.201	-.203	-.013	.154	.260	.366
.350	-.182	-.089	.027	.105	.222	-.177	-.029	.117	.210	.304
.450	-.186	-.112	.001	.068	.217	-.165	-.031	.089	.164	.245
.550	-.154	-.091	.001	.048	.185	-.135	-.031	.068	.132	.194
.650	-.105	-.066	.001	.043	.158	-.095	-.010	.068	.114	.155
.750	-.060	-.031	-.001	.057	.139	-.037	.024	.087	.109	.128
.850	-.007	.013	.017	.078	.130	.028	.073	.112	.119	.110
.900	.012	.015	.020	.078	.119	.054	.084	.114	.116	.087

$\frac{x}{c_v}$	C_p for $z/b_v = 0.66$ β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	C_p for $z/b_v = 0.93$ β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
Left side										
.000	-.335	.578	-.345	-1.082	-1.005	.186	.863	-.359	-.622	-.554
.025	.473	.068	-.784	-1.016	-.980	.821	.193	-.717	-.435	-.598
.075	.357	.059	-.590	-1.034	-.984	.627	.158	-.687	-.414	-.536
.150	.280	.061	-.255	-1.052	-.952	.413	.112	-.551	-.421	-.529
.250	.240	.059	-.140	-.629	-.893	.285	.089	-.401	-.405	-.463
.350	.189	.045	-.131	-.102	-.721					
.450	.131	.015	-.126	-.084	-.540					
.550	.084	.003	-.112	-.118	-.419					
.650	.068	.020	-.061	-.100	-.334	.021	-.008	-.040	-.184	-.302
.750	.068	.050	-.008	-.071	-.284	.000	.006	-.008	-.162	-.286
.850	.068	.073	.038	-.032	-.231	.012	.036	.031	-.134	-.296
.900	.075	.089	.066	.000	-.202	.024	.063	.052	-.114	-.300
Right side										
.025	-.778	.075	.488	.581	.613	-.764	.239	.849	.961	1.009
.075	-.596	.052	.357	.492	.602	-.725	.154	.606	.770	.897
.150	-.247	.066	.292	.419	.535	-.485	.135	.433	.570	.704
.250	-.130	.061	.237	.349	.455					
.350	-.128	.045	.181	.273	.373					
.450	-.114	.017	.130	.207	.286					
.550	-.093	.015	.096	.150	.206					
.650	-.053	.022	.075	.105	.135	-.023	.006	.038	.055	.041
.750	-.011	.043	.068	.075	.071	-.011	.006	.006	-.045	-.094
.850	.061	.093	.098	.078	.029	.047	.063	.036	-.066	-.174
.900	.084	.117	.112	.071	-.005	.070	.084	.047	-.091	-.257

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Continued
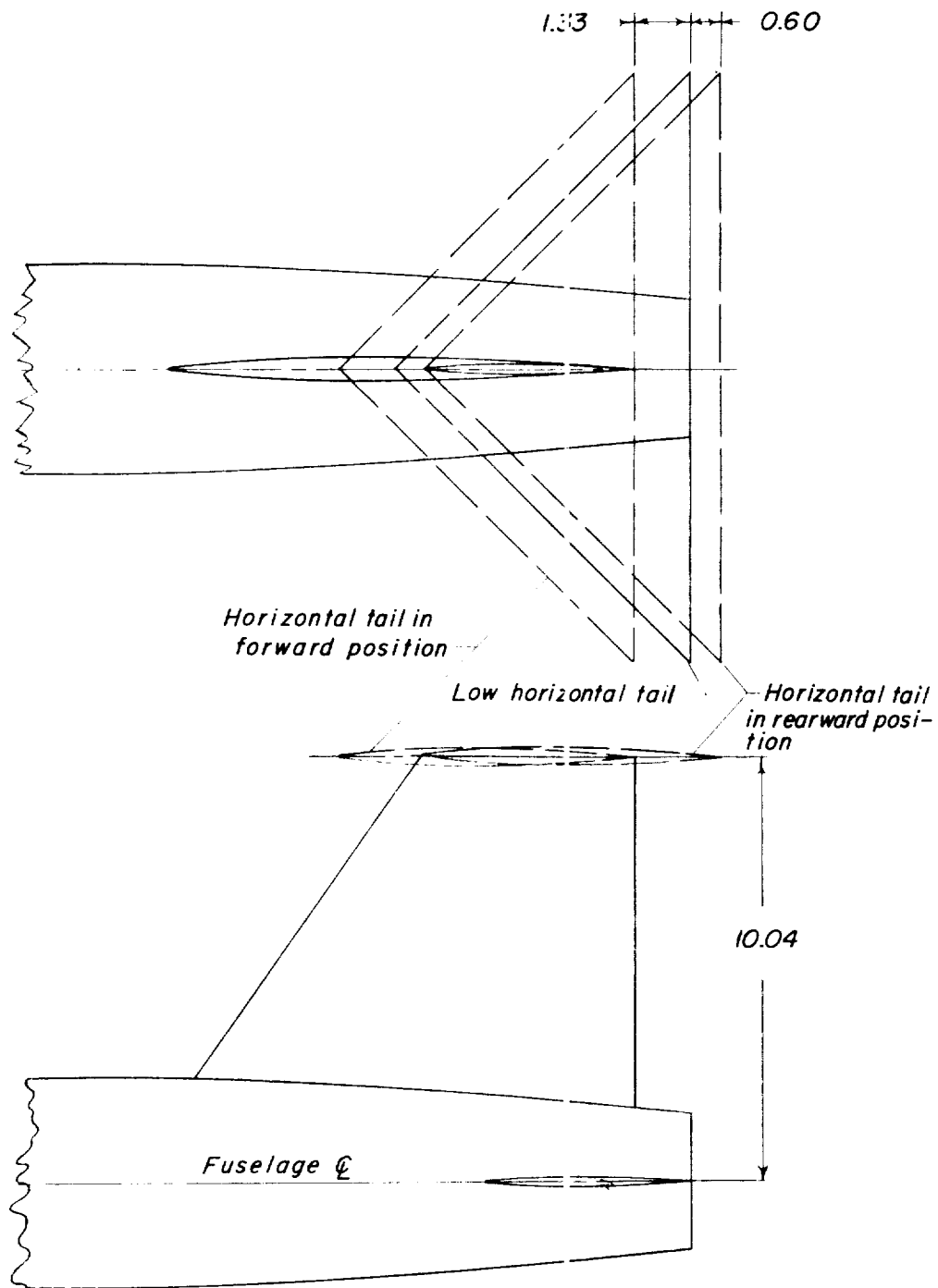
(q) $\alpha = 15.1°$; $M = 0.80$

$\dfrac{x}{c_v}$	C_p for- $z/b_v = 0.11$					C_p for- $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side					Left side				
.000	.574	.905	.816	.714	.412	.110	.841	.206	-.774	-1.499
.025	-.592	-.191	.216	.079	-.248	.417	.056	-.564	-1.505	-1.703
.075	-.159	-.175	-.005	-.100	-.337	.283	.001	-.362	-.677	-1.819
.150	-.035	-.164	-.202	-.274	-.475	.227	.003	-.254	-.490	-.580
.250	.038	-.113	-.188	-.217	-.355	.174	.001	-.204	-.391	-.547
.350	.032	-.071	-.182	-.186	-.274	.129	-.016	-.191	-.343	-.567
.450	.006	-.098	-.202	-.192	-.258	.096	-.033	-.182	-.316	-.482
.550	-.009	-.086	-.167	-.150	-.203	.088	-.025	-.146	-.285	-.421
.650	-.028	-.059	-.112	-.101	-.148	.079	-.008	-.104	-.265	-.375
.750	-.028	-.036	-.069	-.072	-.103	.085	.024	-.043	-.223	-.319
.850	-.003	.010	-.019	-.027	-.031	.105	.073	.019	-.157	-.258
.900	.018	.033	.016	-.001	.016	.122	.097	.061	-.112	-.209
	Right side					Right side				
.025	.320	-.075	-.390	-.018	.178	-.703	.015	.403	.525	.603
.075	-.014	-.191	-.173	.032	.210	-.404	-.023	.267	.402	.541
.150	-.201	-.150	-.025	.084	.245	-.278	-.013	.213	.328	.463
.250	-.197	-.095	.044	.129	.222	-.215	-.005	.177	.272	.397
.350	-.198	-.071	.039	.112	.177	-.195	-.020	.140	.229	.305
.450	-.215	-.100	.021	.079	.189	-.182	-.037	.103	.186	.279
.550	-.171	-.078	.016	.061	.180	-.151	-.036	.088	.150	.232
.650	-.113	-.052	.010	.049	.161	-.107	-.016	.088	.132	.193
.750	-.067	-.026	.015	.059	.145	-.043	.022	.094	.120	.157
.850	-.003	.022	.029	.081	.138	.026	.076	.119	.121	.128
.900	.017	.041	.030	.072	.120	.058	.083	.122	.110	.091

$\dfrac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	Left side					Left side				
.000	-.064	.624	-.065	-.830	-.964	.381	.922	-.034	-.573	-.640
.025	.494	.094	-.846	-1.038	-.933	.858	.192	-.765	-.573	-.669
.075	.376	.079	-.703	-1.053	-.952	.642	.146	-.684	-.524	-.599
.150	.306	.079	-.268	-1.027	-.909	.419	.097	-.610	-.527	-.588
.250	.270	.077	-.139	-.763	-.891	.297	.088	-.489	-.451	-.537
.350	.212	.054	-.150	-.197	-.759					
.450	.148	.018	-.158	-.117	-.599					
.550	.091	-.004	-.133	-.157	-.493					
.650	.067	.006	-.078	-.150	-.426	.003	-.031	-.059	-.274	-.381
.750	.059	.033	-.022	-.123	-.371	-.029	-.016	-.022	-.259	-.366
.850	.058	.065	.033	-.078	-.304	-.041	.024	.021	-.243	-.363
.900	.067	.085	.061	-.055	-.271	-.034	.050	.042	-.232	-.368
	Right side					Right side				
.025	-.810	.108	.520	.612	.652	-.872	.259	.882	1.019	1.086
.075	-.689	.082	.384	.516	.632	-.691	.158	.636	.814	.966
.150	-.305	.096	.323	.454	.577	-.593	.137	.448	.603	.762
.250	-.137	.088	.277	.388	.513					
.350	-.148	.054	.210	.309	.430					
.450	-.142	.030	.154	.235	.345					
.550	-.124	.007	.108	.167	.256					
.650	-.078	.013	.077	.112	.183	-.072	-.017	.021	.027	.085
.750	-.023	.038	.067	.066	.103	-.038	-.008	-.020	-.066	-.042
.850	.053	.096	.094	.056	.048	.023	.061	.004	-.107	-.122
.900	.081	.114	.099	.039	.001	.045	.079	.000	-.188	-.224

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION; $i_t = 0°$ - Concluded

(r) $\alpha = 15.1°$; M = 0.85

$\dfrac{x}{c_v}$	C_p for- $z/b_v = 0.11$					C_p for- $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.637	.928	.851	.756	.417	.197	.868	.295	-.570	-1.255
.025	-.650	-.200	.225	.123	-.174	.410	.043	-.538	-1.351	-1.577
.075	-.166	-.195	.008	-.065	-.272	.284	-.019	-.376	-.706	-1.623
.150	-.035	-.179	-.208	-.280	-.450	.227	-.014	-.271	-.541	-.741
.250	.033	-.129	-.199	-.249	-.402	.177	-.013	-.215	-.427	-.422
.350	.030	-.087	-.194	-.220	-.296	.132	-.032	-.205	-.390	-.644
.450	.005	-.110	-.204	-.211	-.236	.095	-.054	-.197	-.358	-.503
.550	-.009	-.102	-.179	-.180	-.232	.088	-.046	-.165	-.327	-.467
.650	-.027	-.082	-.126	-.127	-.176	.080	-.027	-.121	-.304	-.417
.750	-.035	-.053	-.083	-.098	-.141	.080	.001	-.059	-.251	-.355
.850	-.007	-.003	-.028	-.043	-.059	.102	.054	.010	-.179	-.285
.900	.013	.020	.010	-.012	-.008	.115	.079	.054	-.122	-.231
Right side										
.025	.343	-.083	-.430	-.043	.165	-.694	.000	.394	.518	.611
.075	.005	-.216	-.178	.021	.190	-.413	-.044	.265	.405	.541
.150	-.209	-.163	-.028	.082	.232	-.293	-.027	.213	.331	.467
.250	-.211	-.110	.046	.128	.227	-.229	-.022	.179	.283	.404
.350	-.209	-.084	.041	.113	.170	-.204	-.039	.139	.207	.280
.450	-.226	-.112	.018	.082	.167	-.198	-.060	.110	.197	.290
.550	-.185	-.094	.014	.062	.169	-.168	-.056	.090	.162	.243
.650	-.119	-.070	.010	.053	.157	-.118	-.032	.087	.143	.202
.750	-.075	-.040	.014	.058	.142	-.054	.004	.096	.128	.185
.850	-.012	.010	.025	.078	.133	.020	.064	.116	.125	.132
.900	.013	.029	.033	.075	.117	.053	.066	.121	.109	.092

$\dfrac{x}{c_v}$	C_p for- $z/b_v = 0.66$					C_p for- $z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	-.012	.634	.605	-.716	-.901	.440	.936	.081	-.606	-.716
.025	.503	.086	-.866	-1.105	-.884	.869	.194	-.750	-.714	-.701
.075	.384	.074	-.748	-1.113	-.912	.654	.124	-.707	-.672	-.646
.150	.320	.074	-.298	-1.033	-.897	.431	.083	-.632	-.682	-.644
.250	.279	.067	-.146	-.736	-.890	.306	.071	-.526	-.626	-.606
.350	.219	.041	-.165	-.377	-.777					
.450	.152	.001	-.179	-.220	-.647					
.550	.090	-.023	-.158	-.211	-.552					
.650	.066	-.017	-.098	-.192	-.487	-.002	-.064	-.081	-.321	-.450
.750	.052	.010	-.039	-.150	-.426	-.041	-.047	-.050	-.291	-.427
.850	.048	.043	.020	-.095	-.352	-.064	-.006	-.002	-.271	-.415
.900	.059	.060	.051	-.068	-.318	-.068	.023	.021	-.266	-.412
Right side										
.025	-.873	.100	.530	.620	.667	-.885	.255	.894	1.041	1.109
.075	-.751	.076	.388	.525	.640	-.699	.136	.643	.834	.978
.150	-.320	.089	.332	.464	.587	-.621	.123	.452	.623	.779
.250	-.145	.080	.288	.400	.523					
.350	-.165	.047	.220	.323	.441					
.450	-.166	.009	.162	.251	.360					
.550	-.138	-.014	.110	.179	.272					
.650	-.095	-.010	.077	.125	.199	-.091	-.050	.017	.052	.103
.750	-.037	.014	.063	.078	.119	-.061	-.040	-.033	-.049	-.017
.850	.046	.077	.091	.062	.059	-.001	.031	-.016	-.086	-.087
.900	.073	.094	.091	.041	-.001	.025	.049	-.029	-.169	-.185
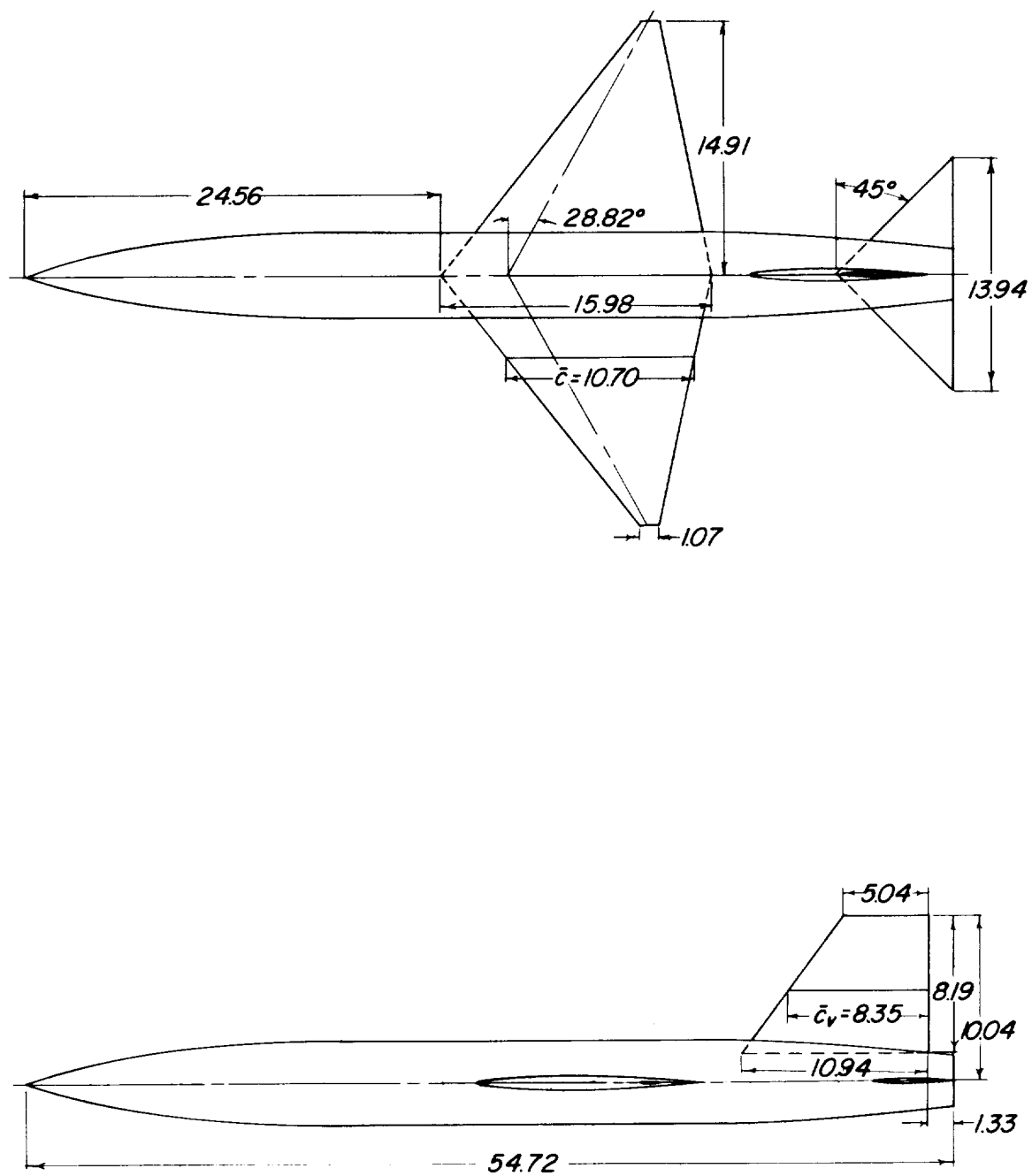
(a) Complete model with horizontal tail in low position.

Figure 1.- Details of test model. (All linear dimensions are in inches.)

1.33   0.60

Horizontal tail in
forward position

Low horizontal tail

Horizontal tail
in rearward posi-
tion

10.04

Fuselage ₵

(b) Horizontal-tail positions.

Figure 1.- Continued.

Station 4 $c_v = 5.43$

Station 3 $c_v = 7.05$

Station 2 $c_v = 8.67$

Station 1 $c_v = 10.29$

5.04

8.19

7.63

5.38

3.13

0.88

1.85

10.94

Station	$\dfrac{z}{b_v}$
1	0.11
2	0.38
3	0.66
4	0.93

Chordwise location of static orifices, x/c_v	Station 4	
Stations 1,2,3 Left and right sides	Left side	Right side
.000	.000	.000
.025	.025	.025
.075	.075	.075
.150	.150	.150
.250	.250	—
.350	—	—
.450	—	—
.550	—	—
.650	.650	.650
.750	.750	.750
.850	.850	.850
.900	.900	.900
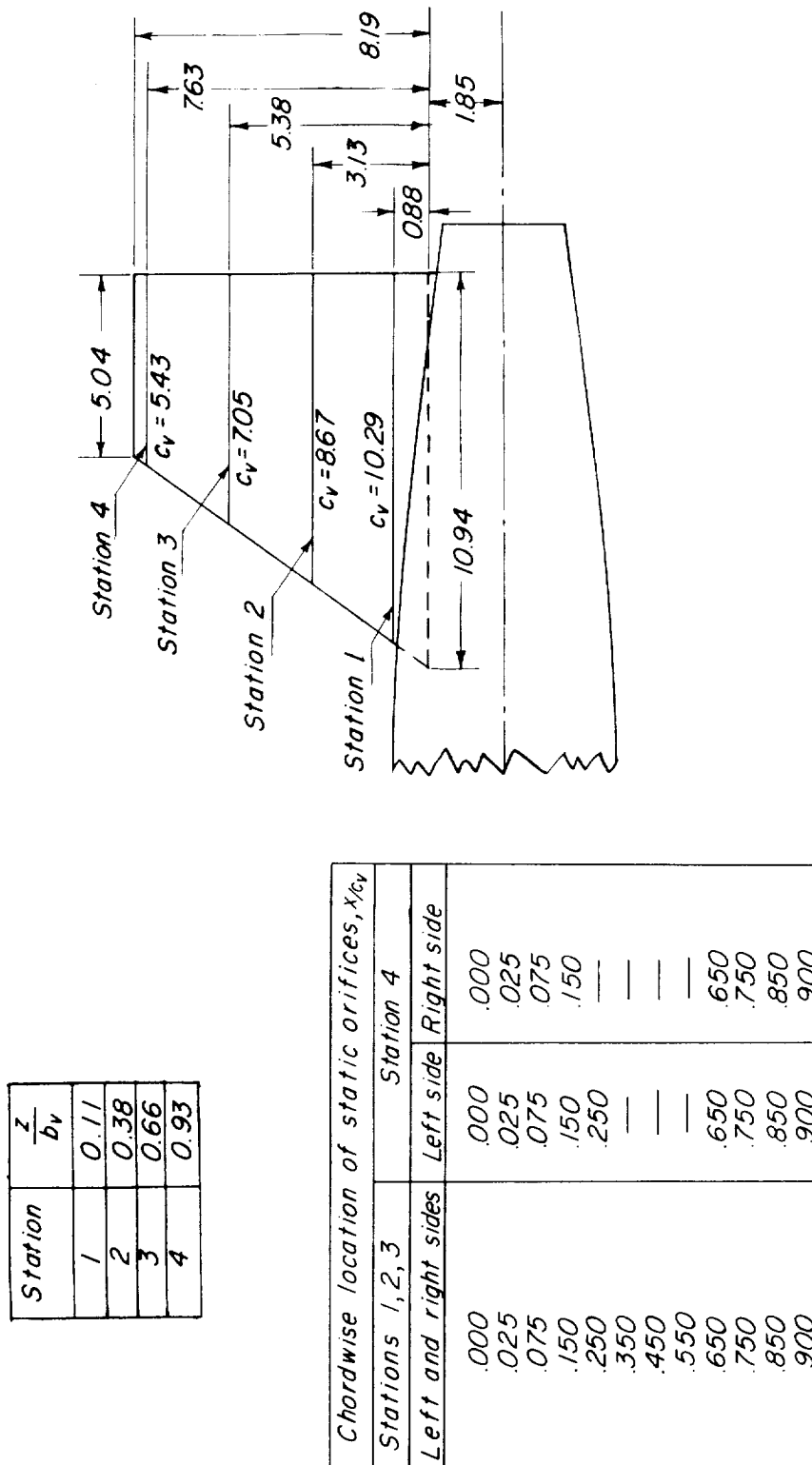
(c) Vertical-tail pressure-orifice locations.

Figure 1.- Continued.
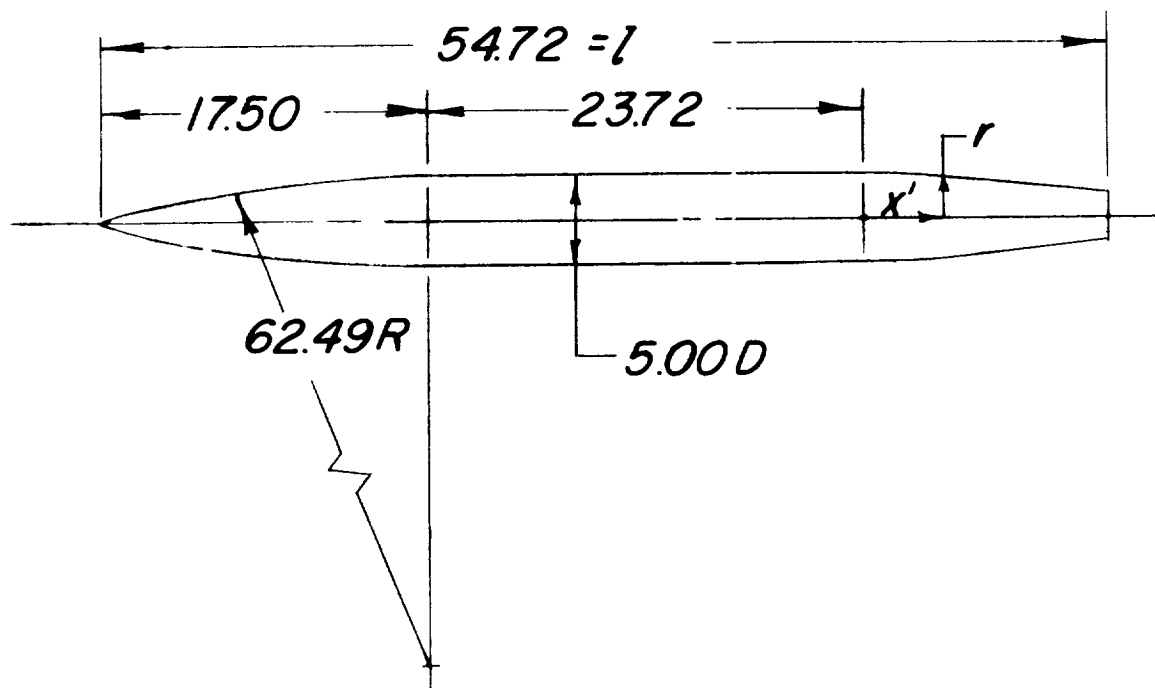
Afterbody Coordinates

x'/l	r/l
0	.0456
.0320	.0445
.0639	.0427
.1187	.0390
Straight-line taper	
.2460	.0301

(d) Fuselage dimensions; fineness ratio, 10.94.

Figure 1.- Concluded.
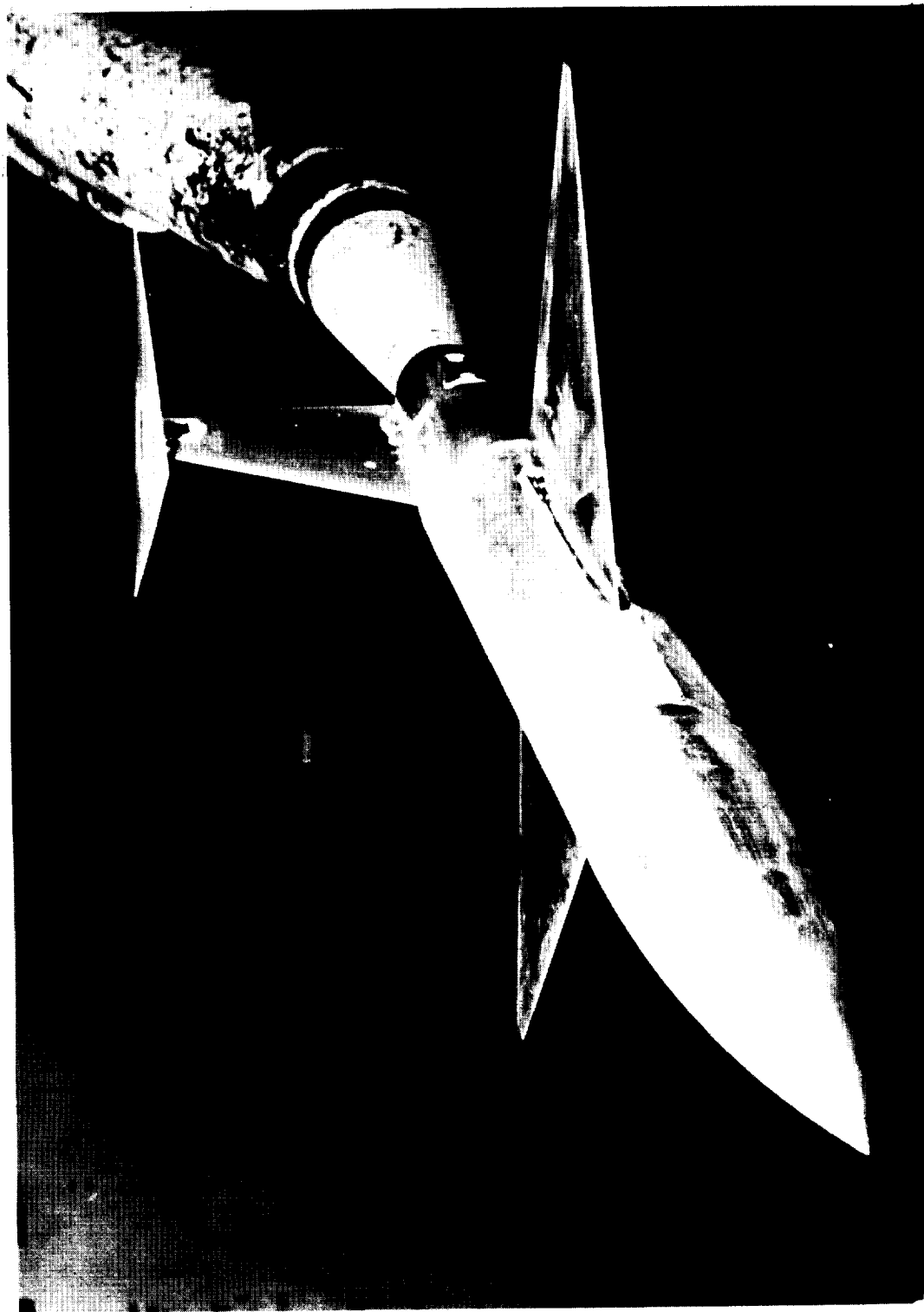
L-89250

Figure 2.- Model mounted in tunnel.
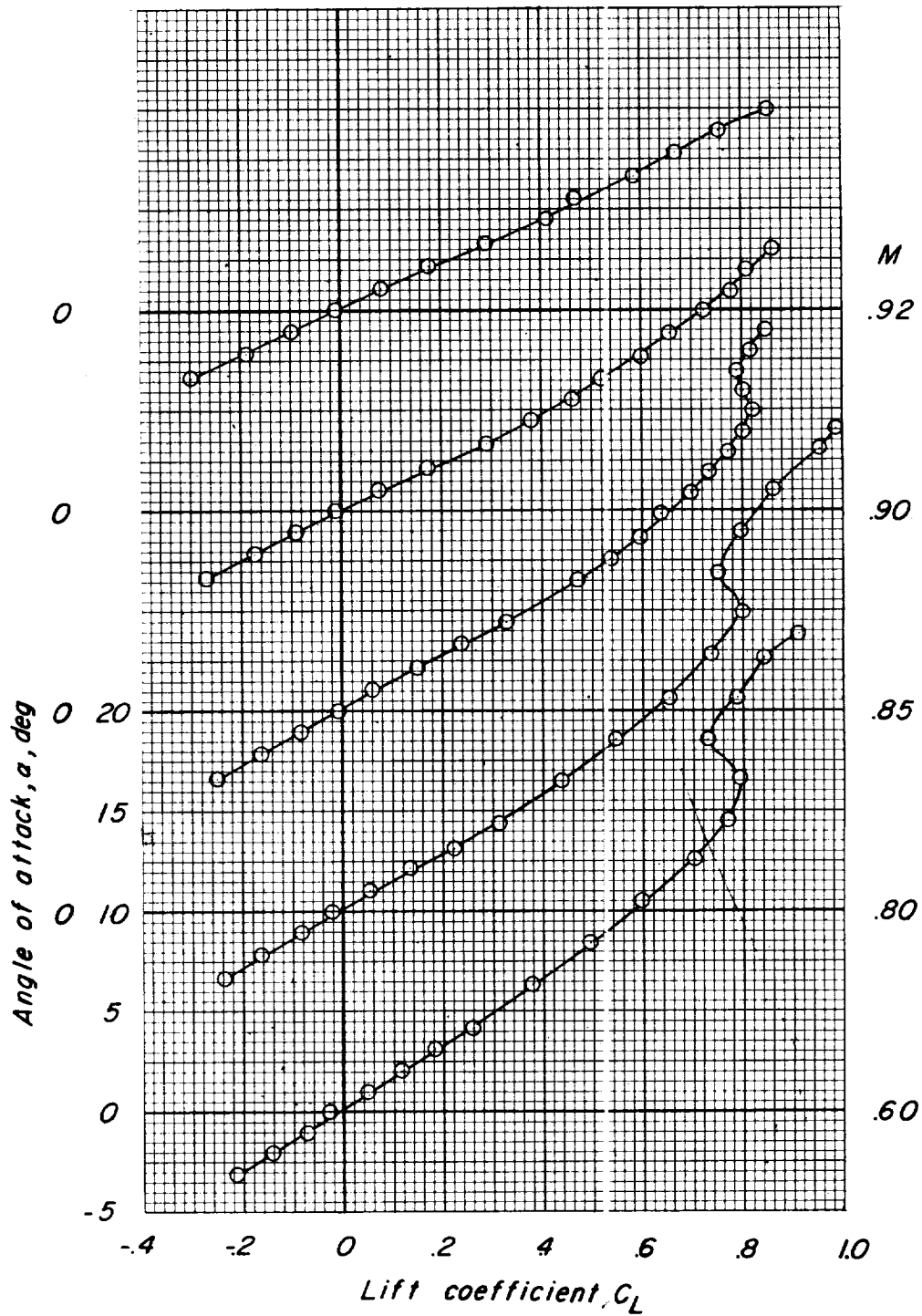
Figure 3.- Variation of model lift coefficient with angle of attack. Horizontal tail off. (Data from ref. 10.)
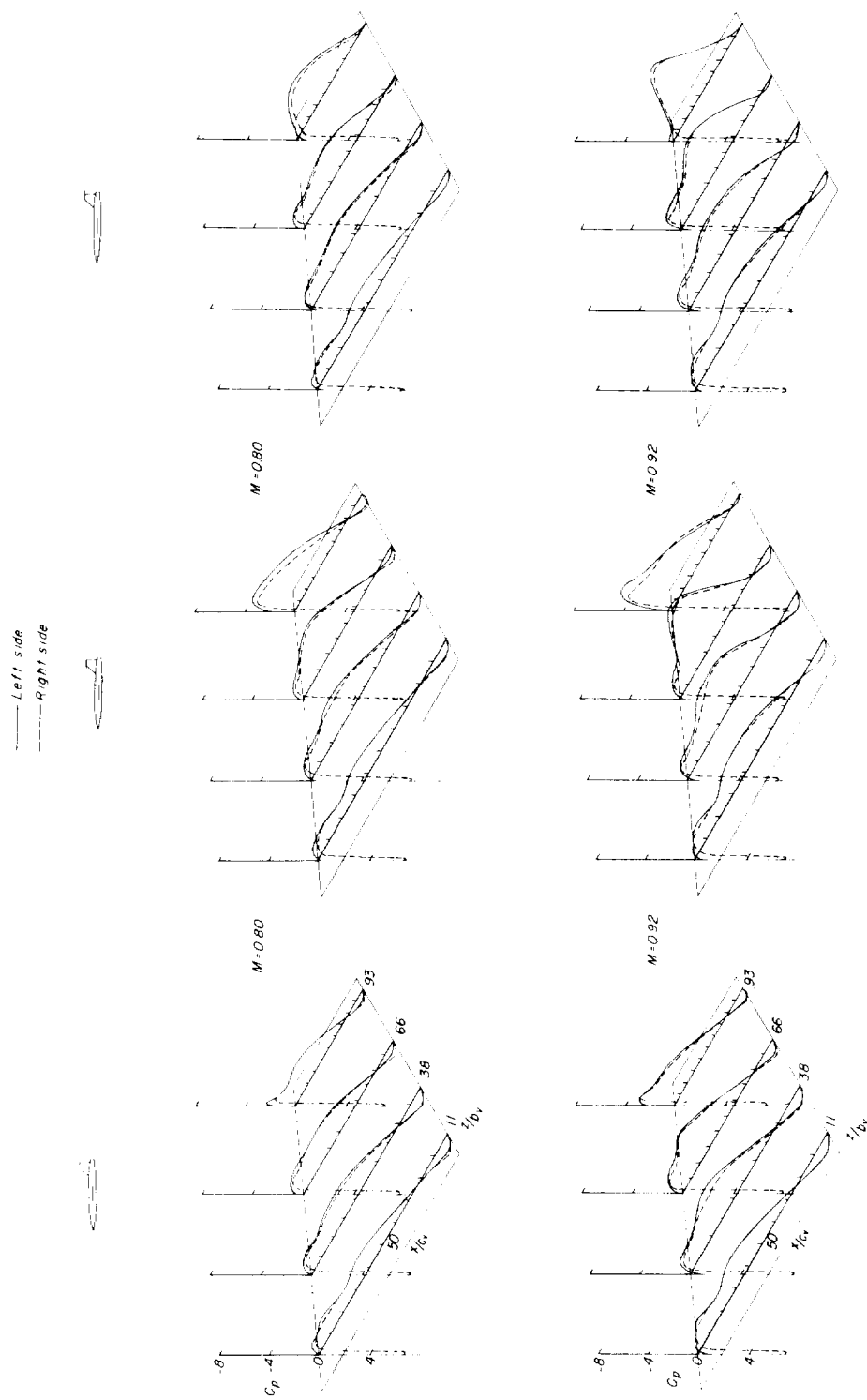
Figure 4.- Effect of horizontal-tail position on vertical-tail pressure distribution. $i_t = 0^\circ$;

(a) $\beta = 0^\circ$.
$\alpha = 0^\circ$.

(b) β = 3.9°.

Figure 4.- Continued.

Left side
Right side

M=0.80

M=0.92

M=0.80

M=0.92

C_p

-16
-12
-8
-4
0
4

93
66
38
11

.50
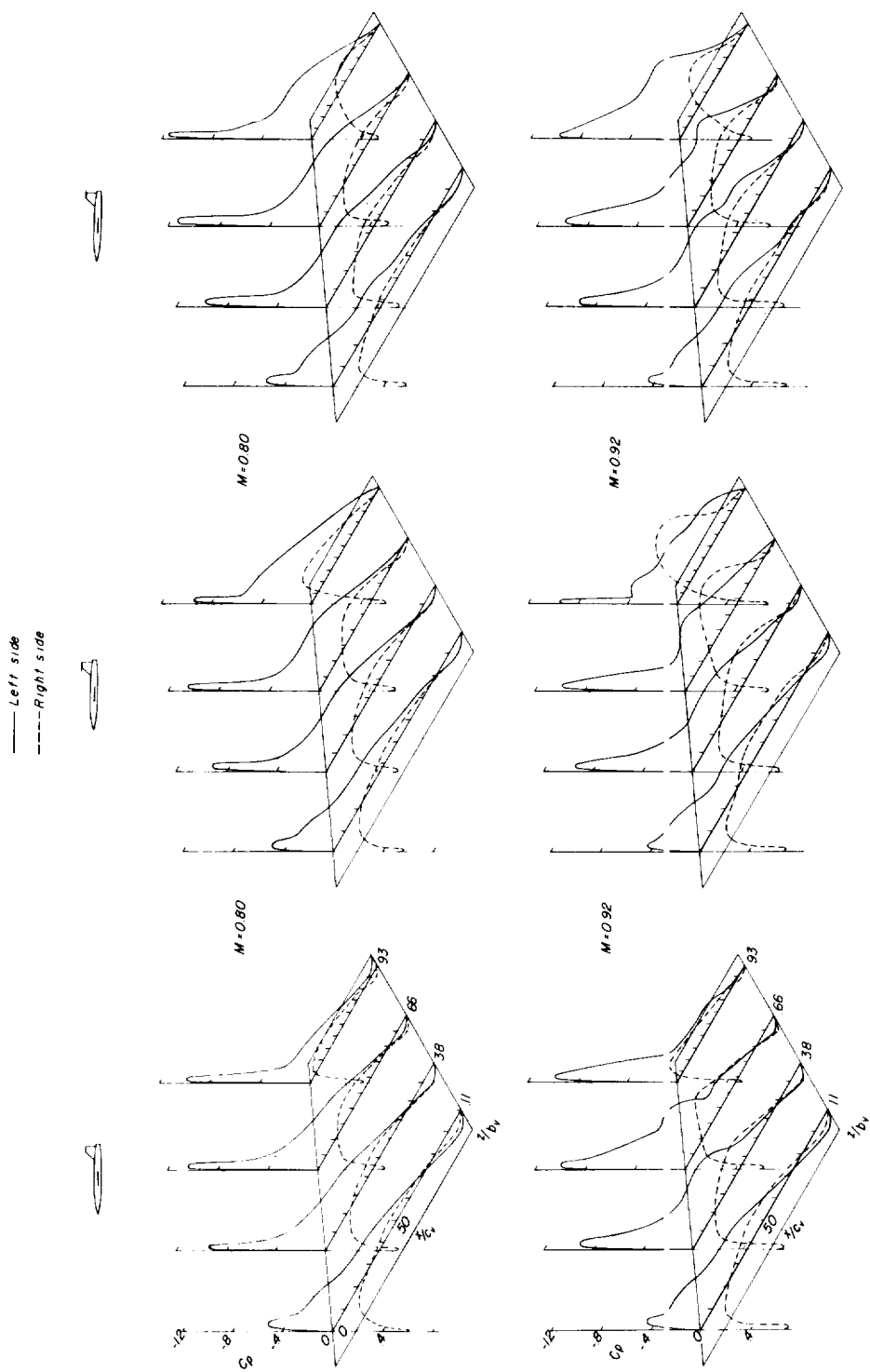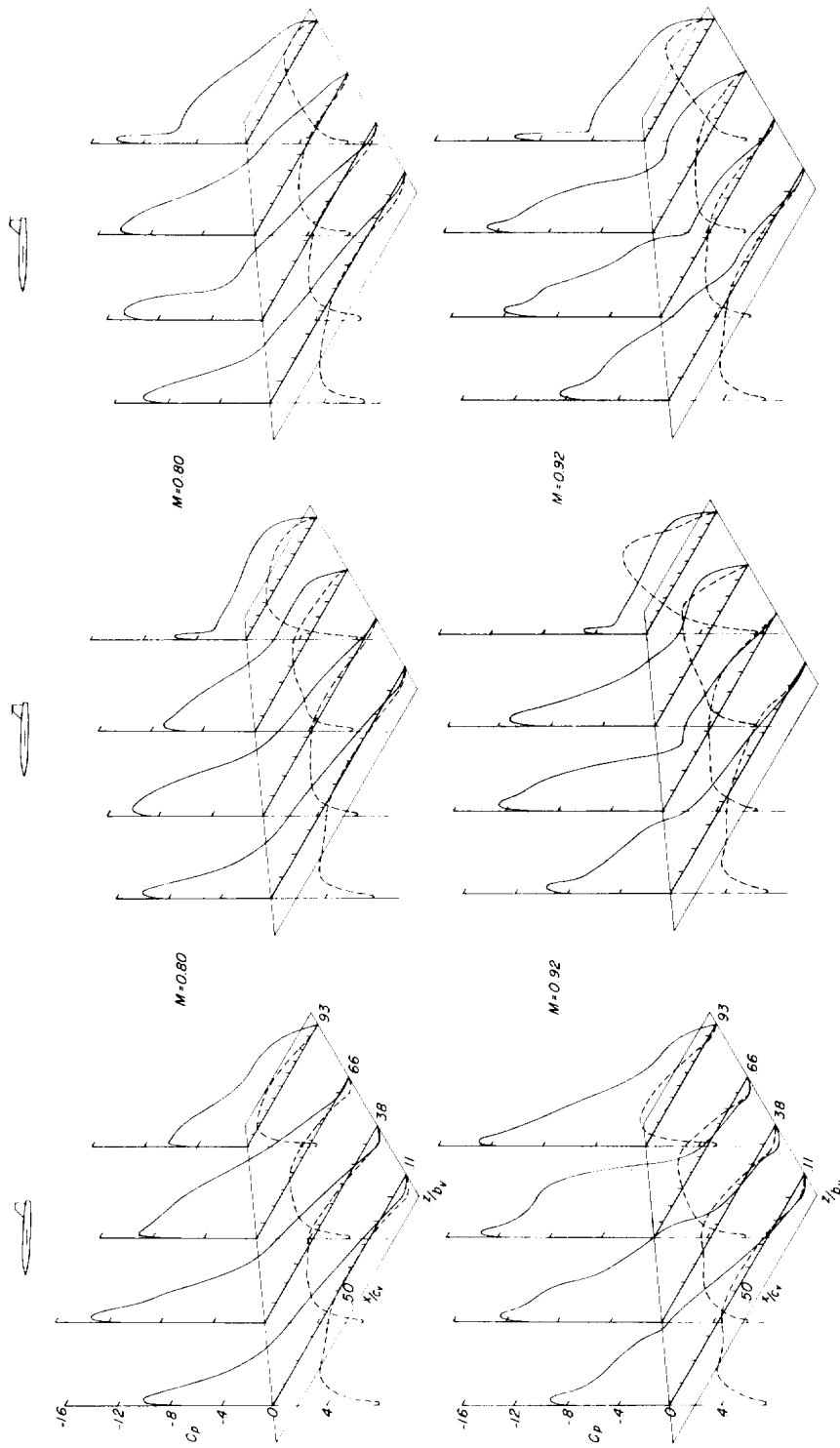
(c) β = 7.8°.

Figure 4.- Continued.
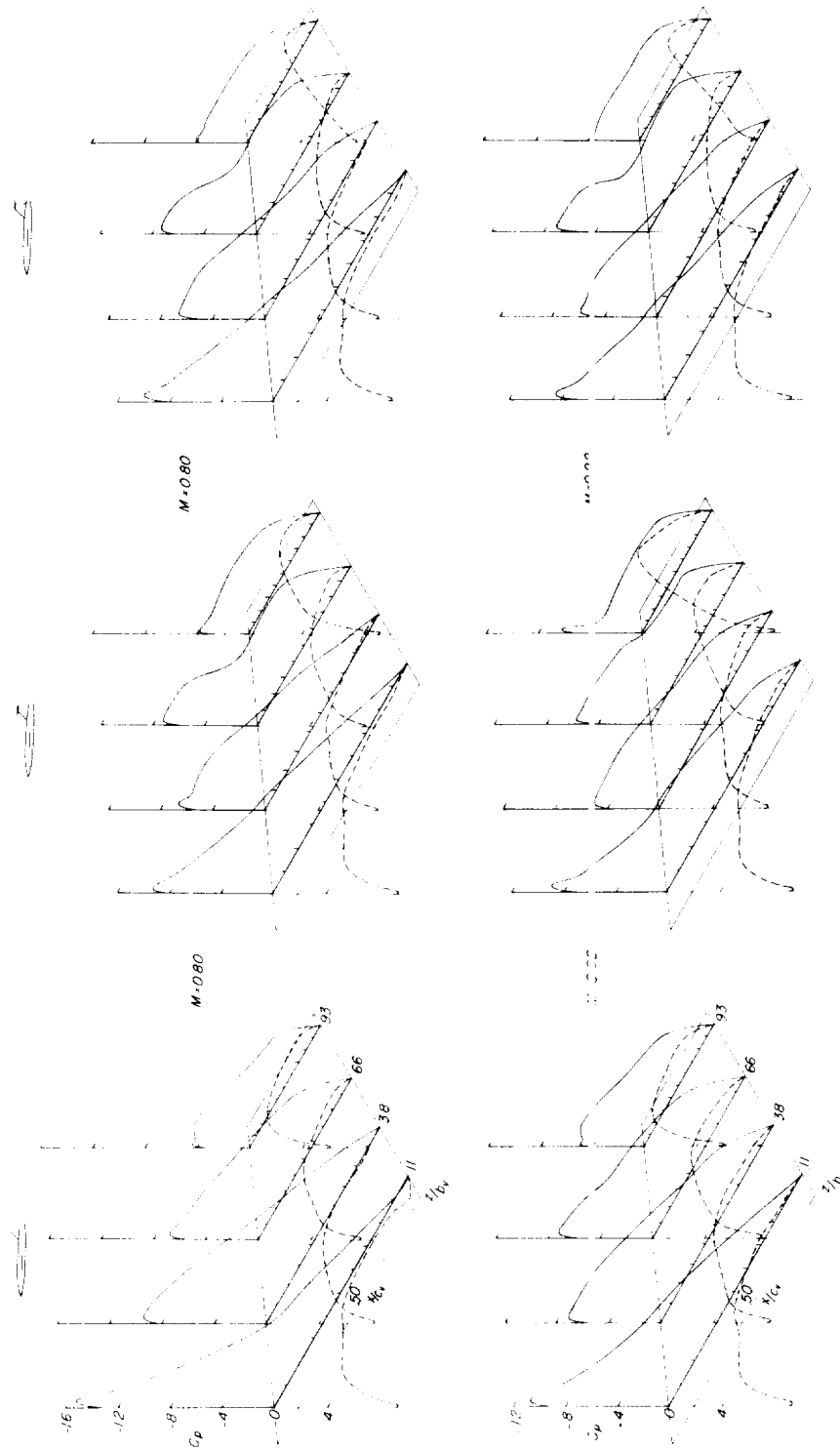
(d) $\beta = 12.7^\circ$.

Figure 4.- Concluded.
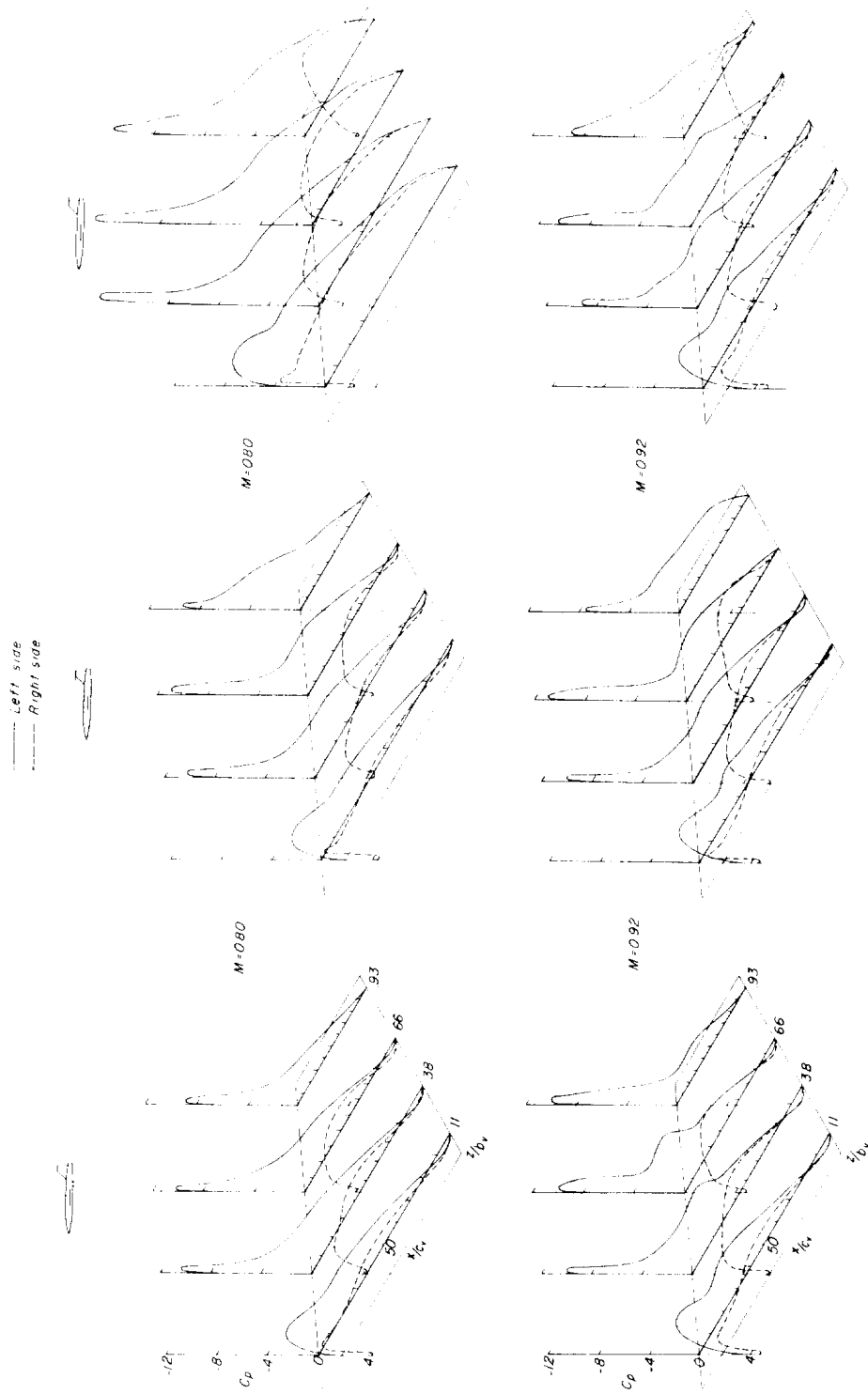
Figure 5.- Effect of horizontal-tail position on vertical-tail pressure distribution. $i_t = 0°$; $\alpha = 5.6°$; $\beta = 3.9°$.

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(a) M = 0.80.
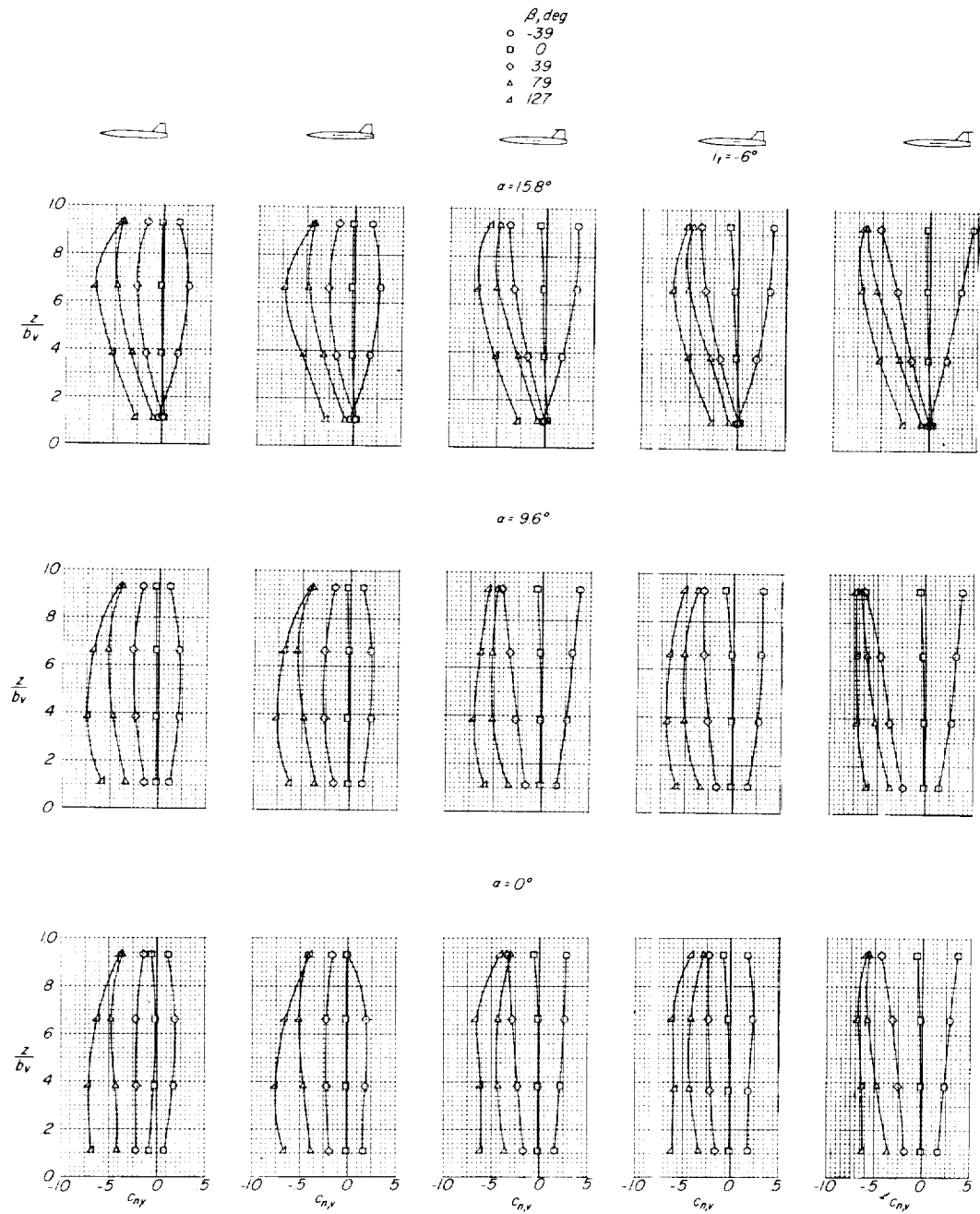
Figure 6.- Effect of sideslip angle on the spanwise variation of
vertical-tail section normal-force coefficient. $i_t = 0°$ (except
where noted).
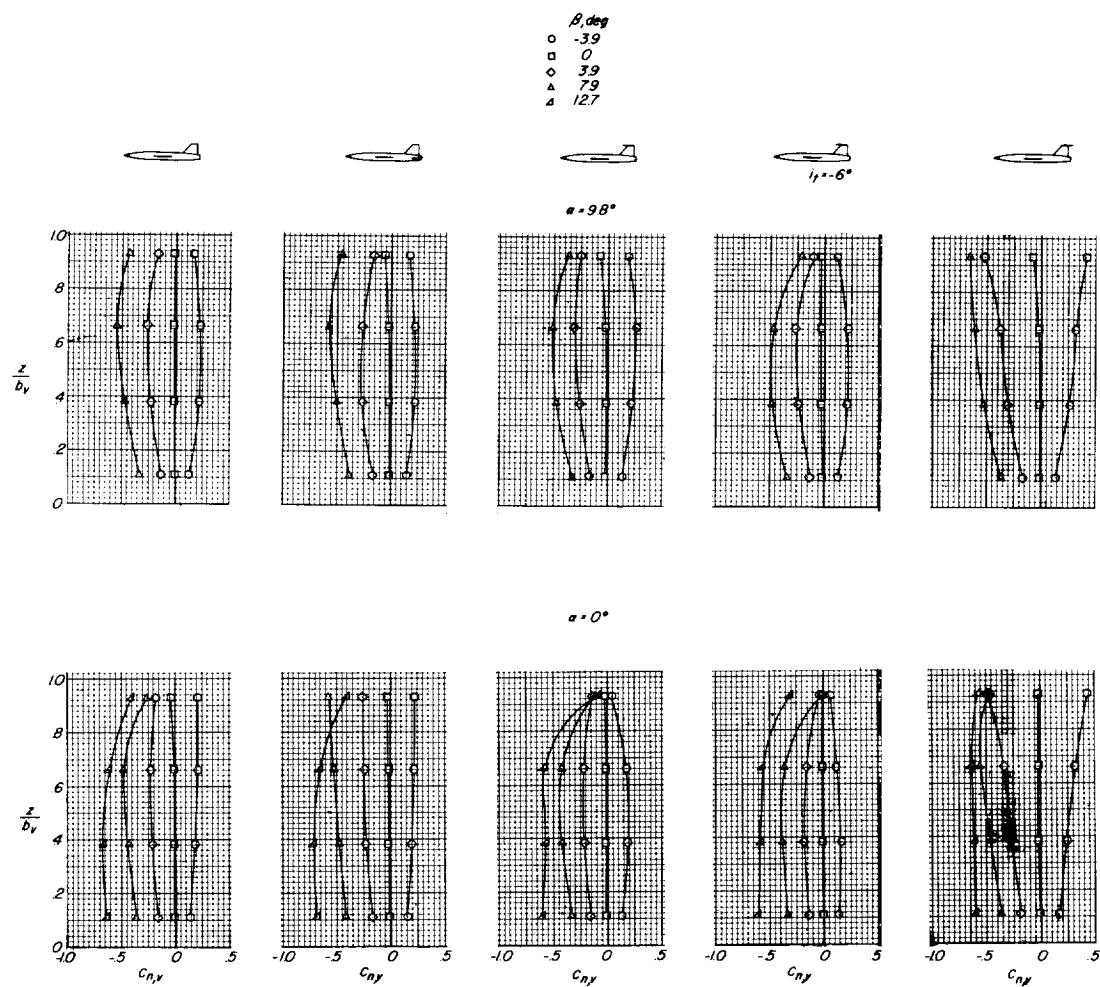
(b)   M = 0.92.

Figure 6.- Concluded.
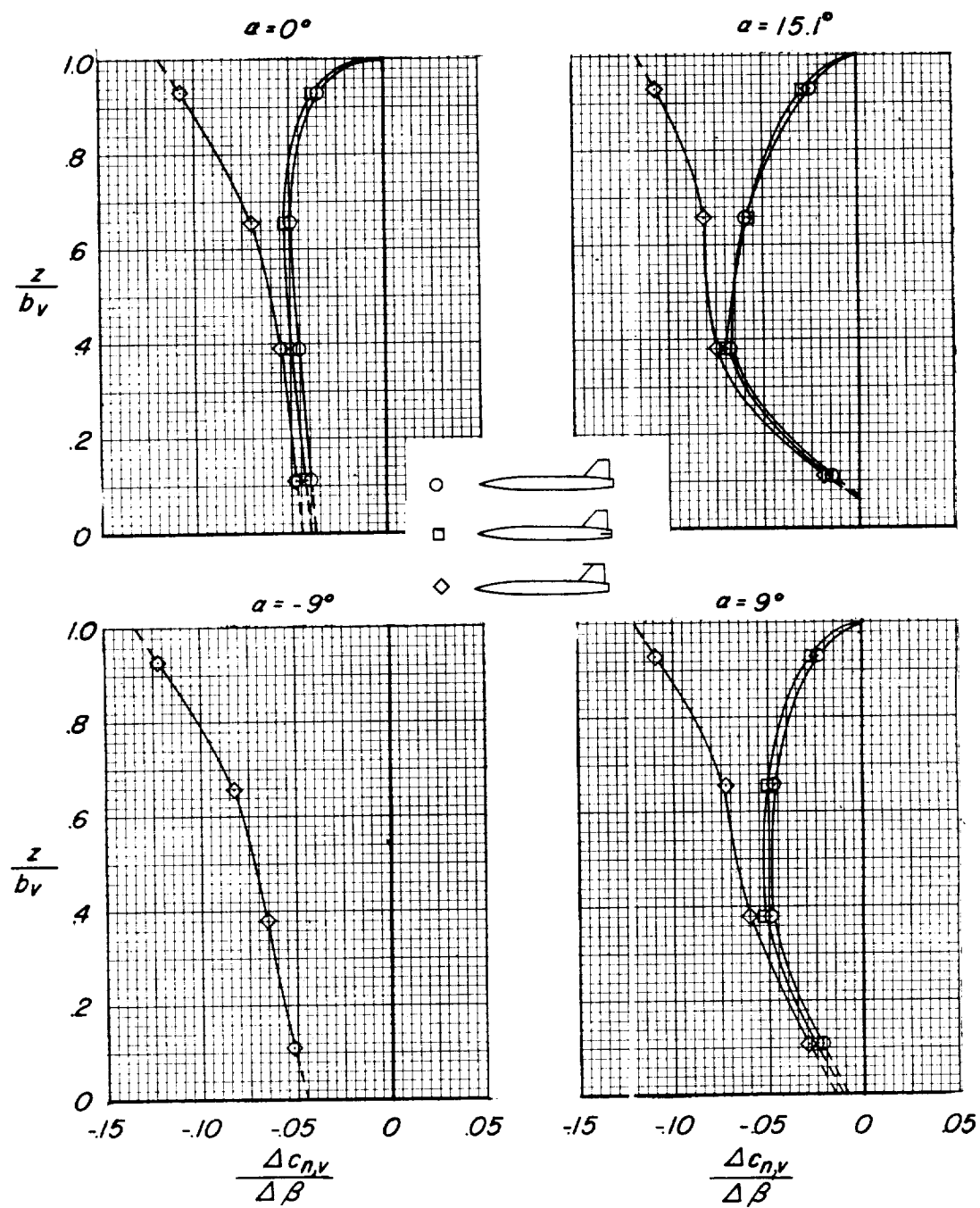
(a) M = 0.60.

Figure 7.- Effect of horizontal-tail position on the spanwise variation
of vertical-tail section normal-force coefficient per degree of
sideslip. Wing-off configuration; $\Delta\beta \approx 3°$; $i_t = 0°$.
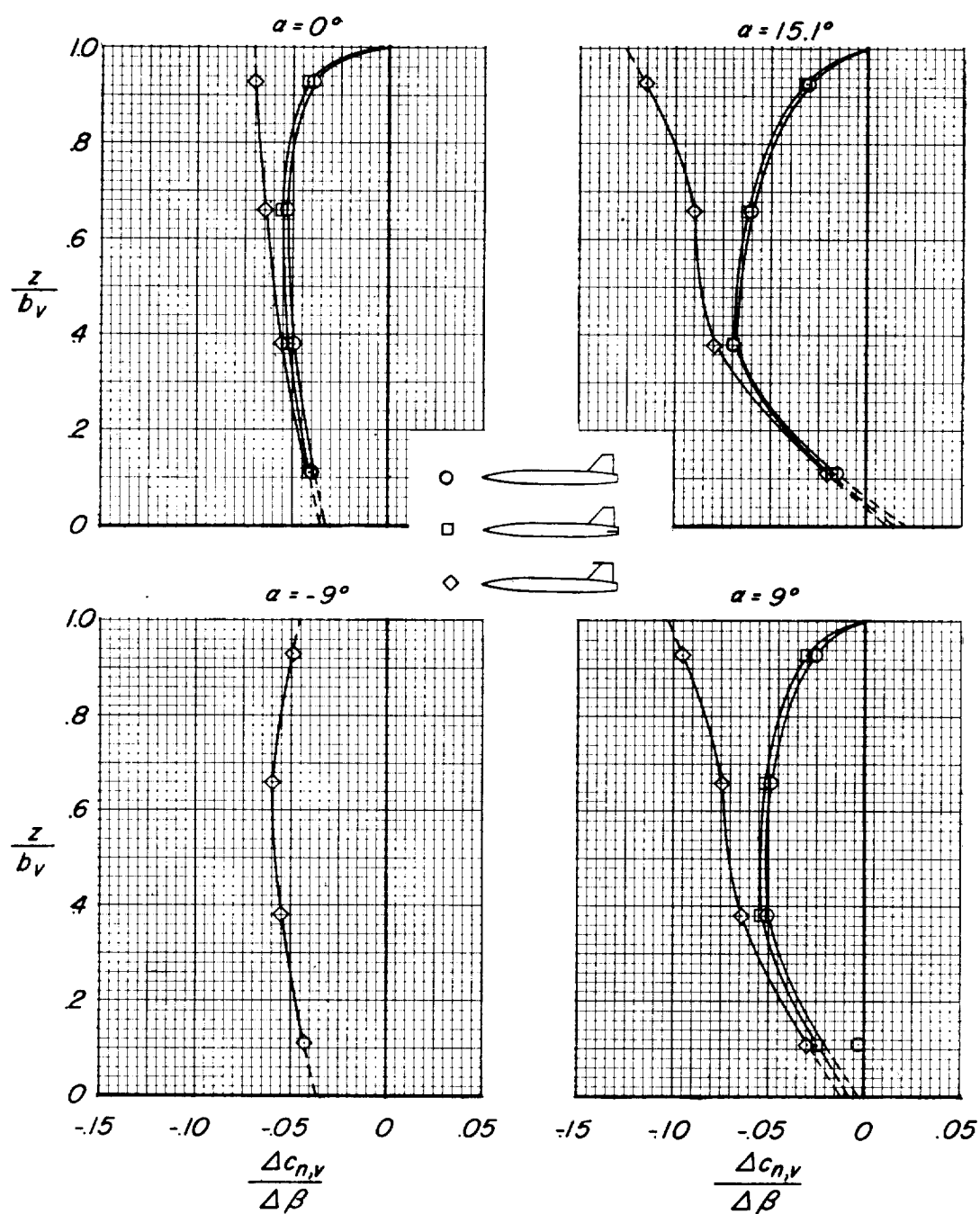
(b)   M = 0.80.

Figure 7.- Continued.

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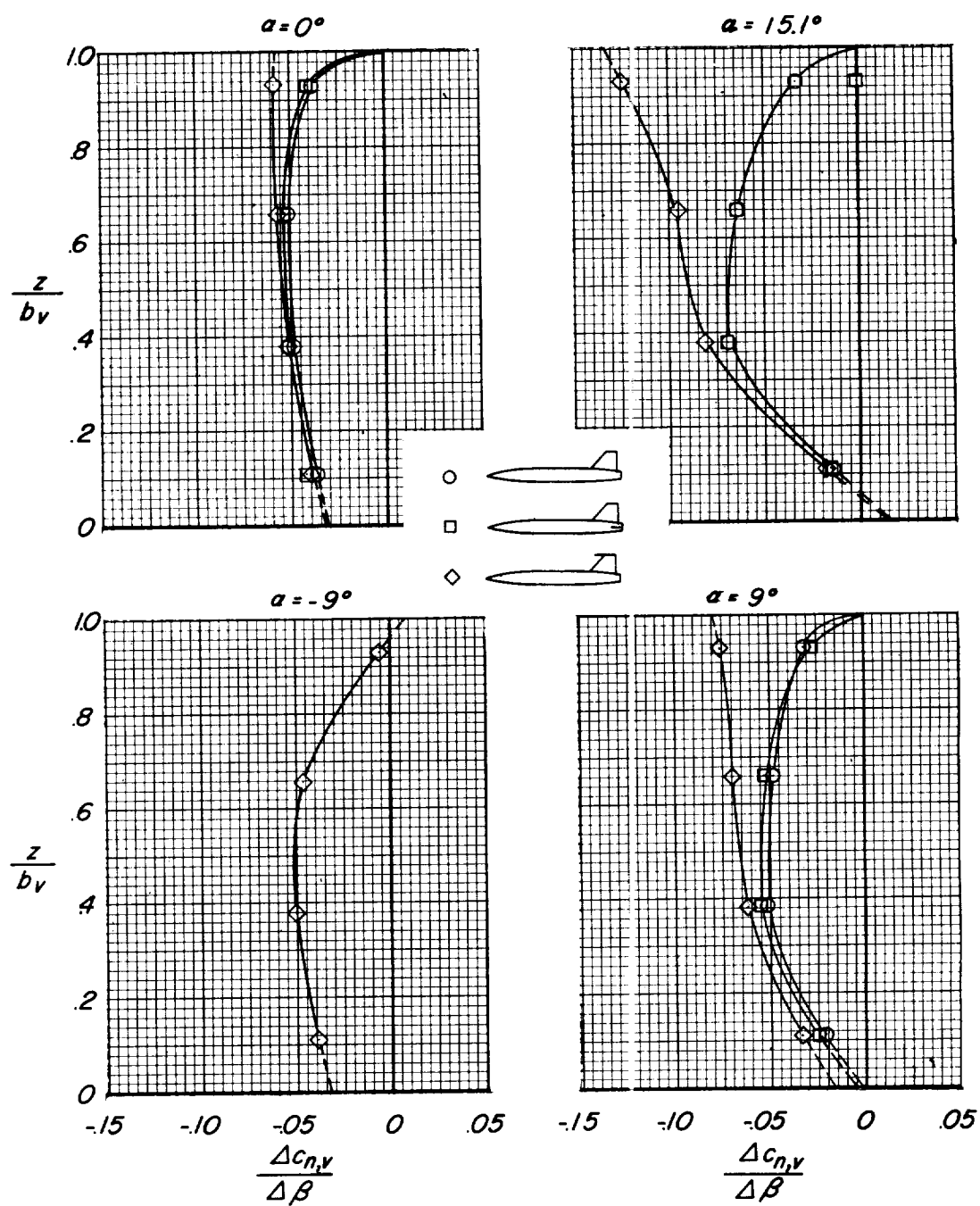
(c) M = 0.85.

Figure 7.- Continued.
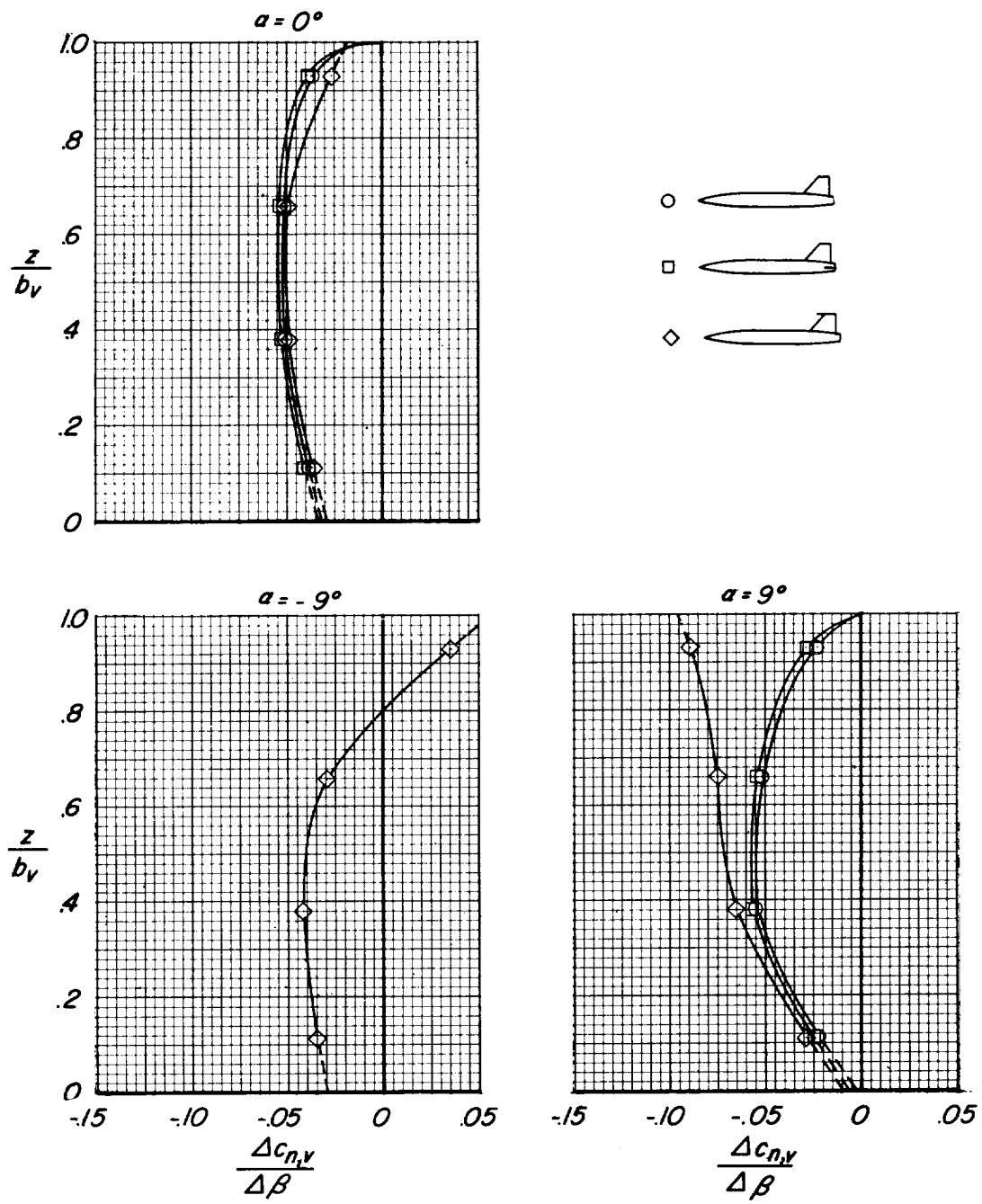
(d)   M = 0.90.

Figure 7.- Continued.
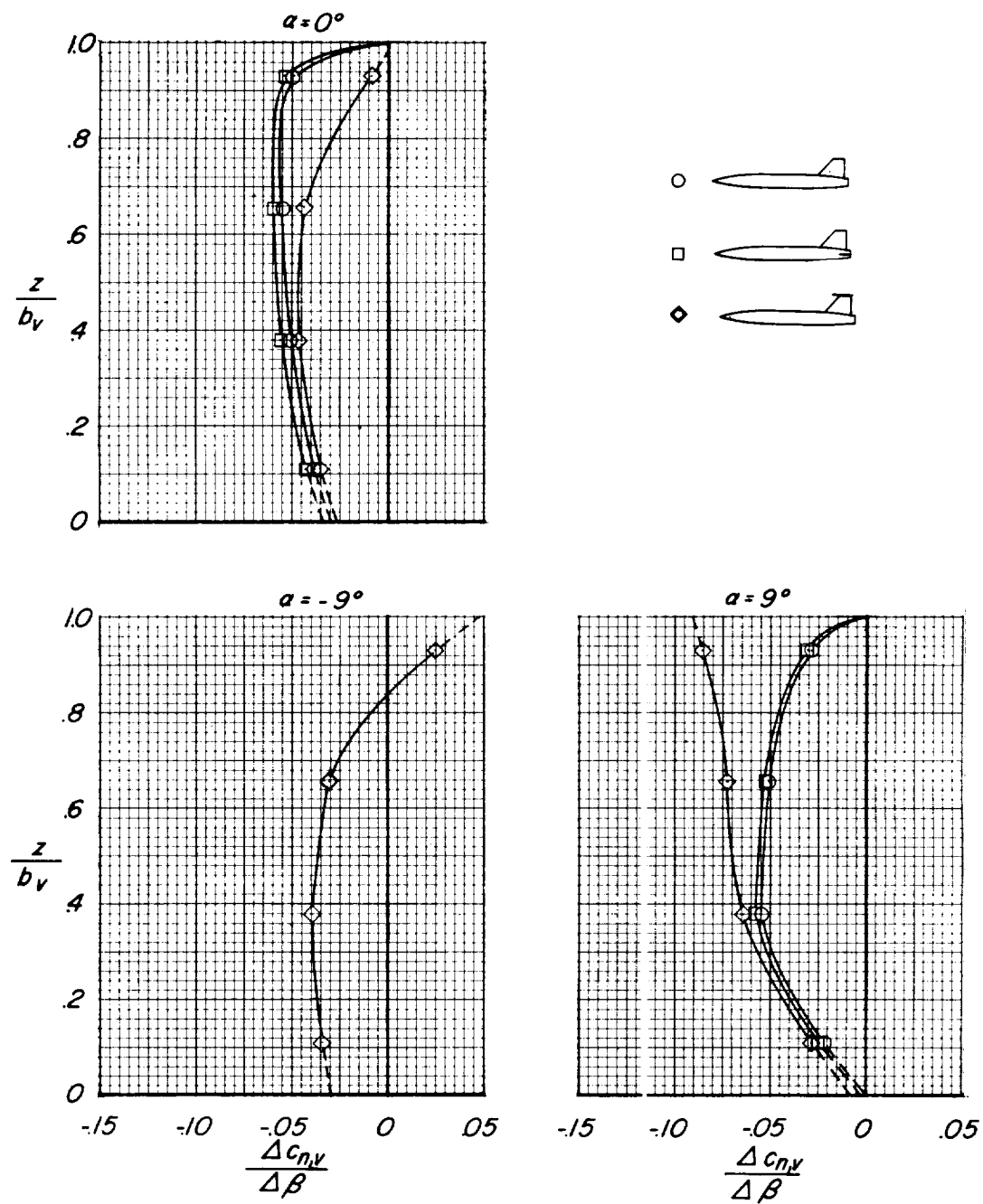
(e)   M = 0.92.

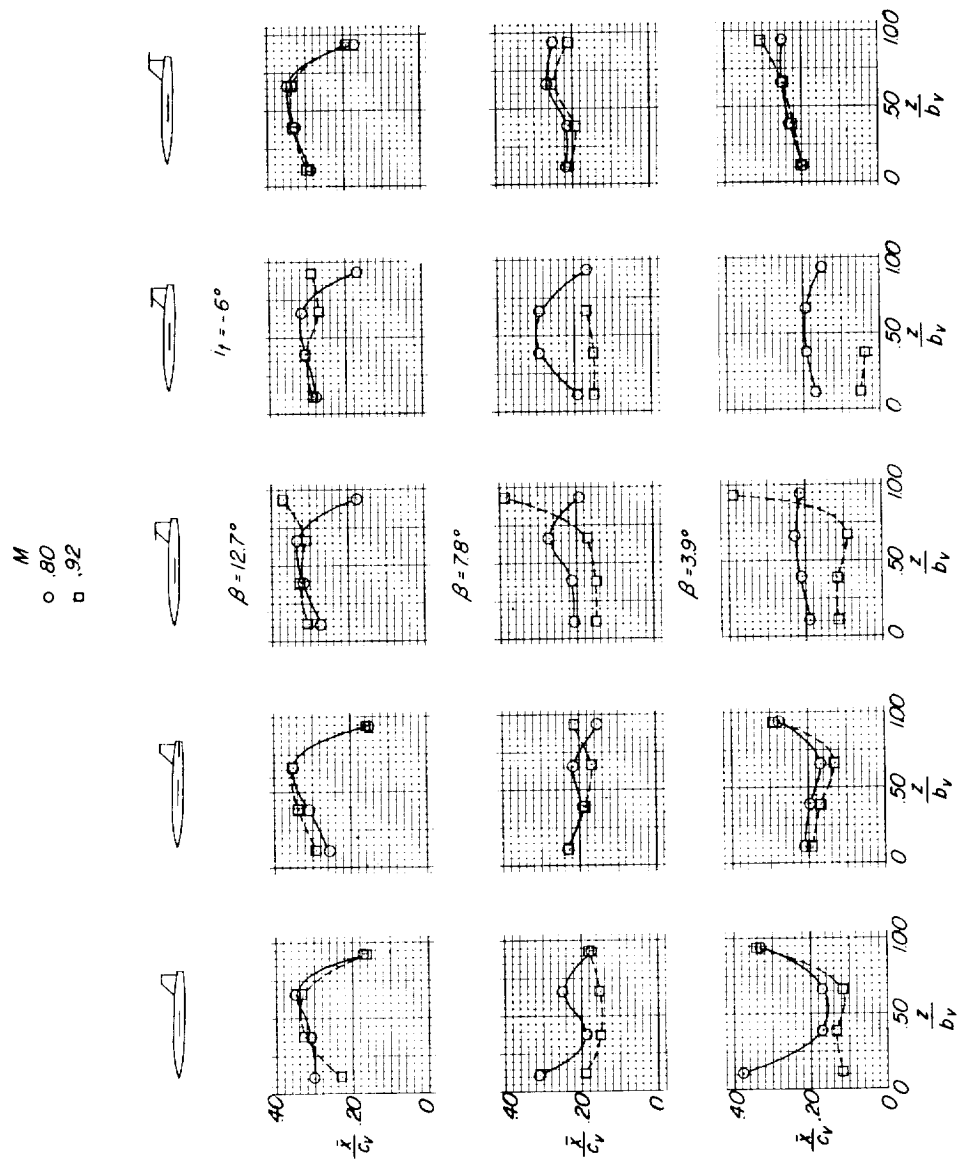Figure 7.- Concluded.
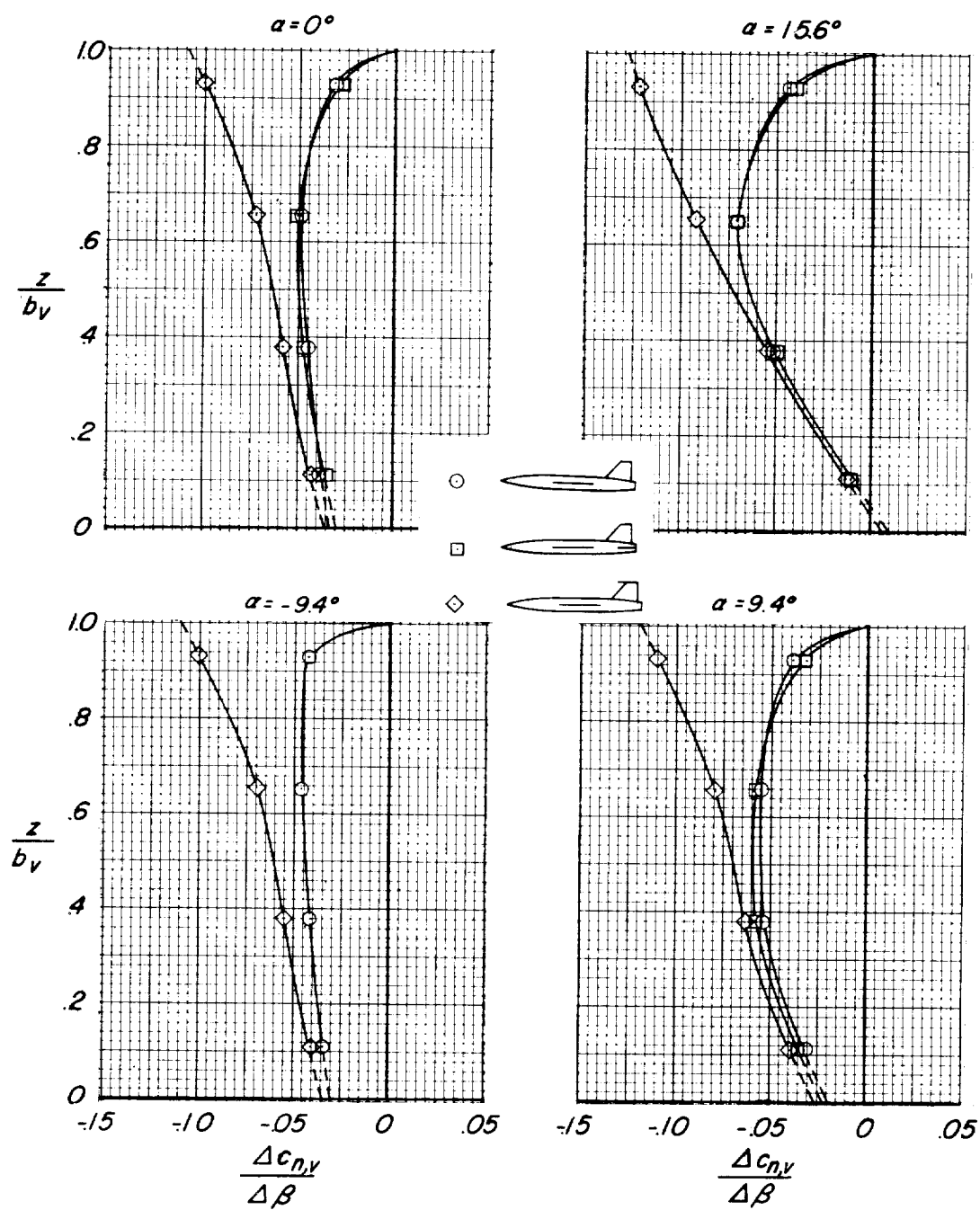
(a) M = 0.60.

Figure 8.- Effect of horizontal-tail position on the spanwise variation
of vertical-tail section normal-force coefficient per degree of
sideslip. $\Delta\beta \approx 8°$; $i_t = 0°$.
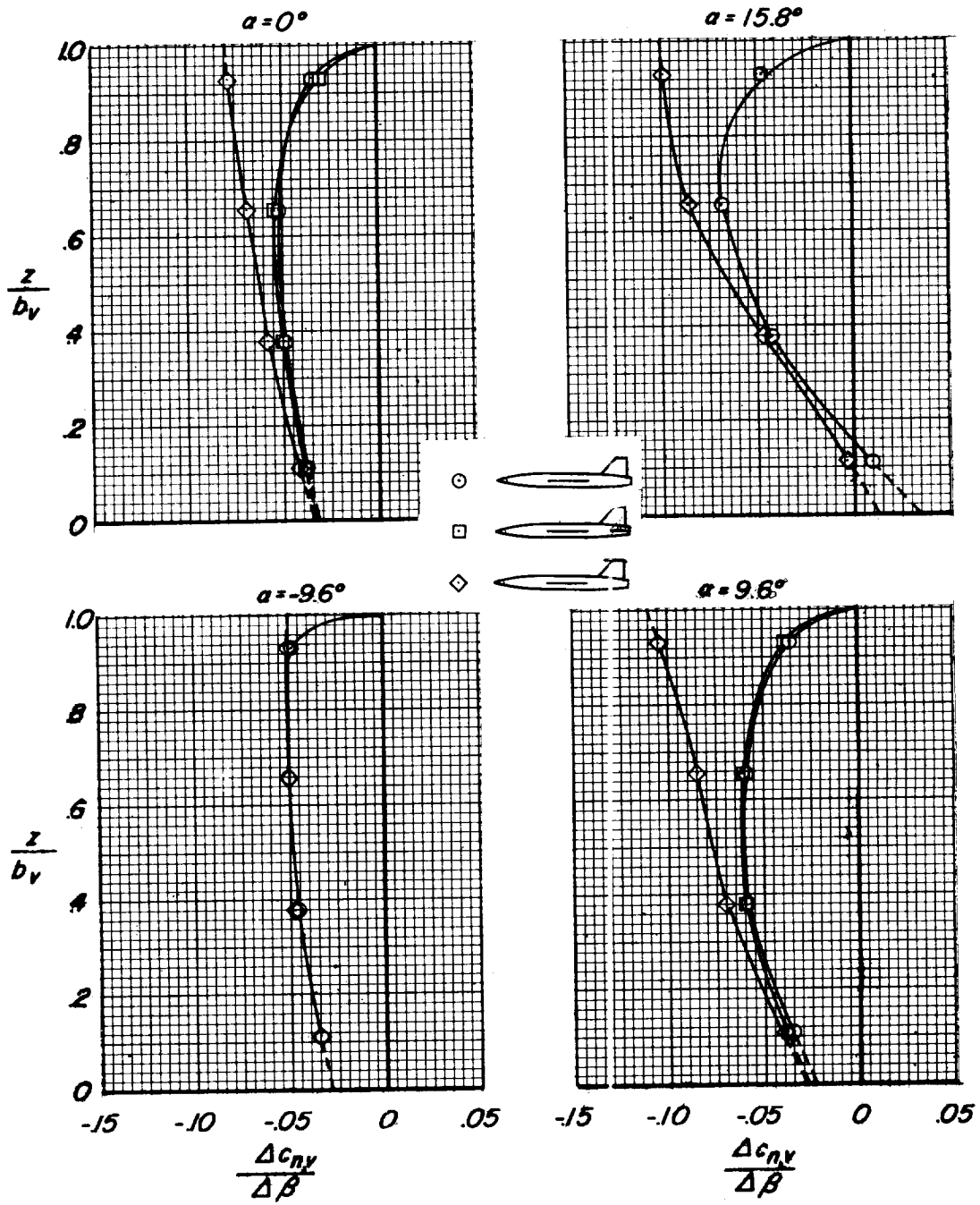
(b) M = 0.80.

Figure 8.- Continued.
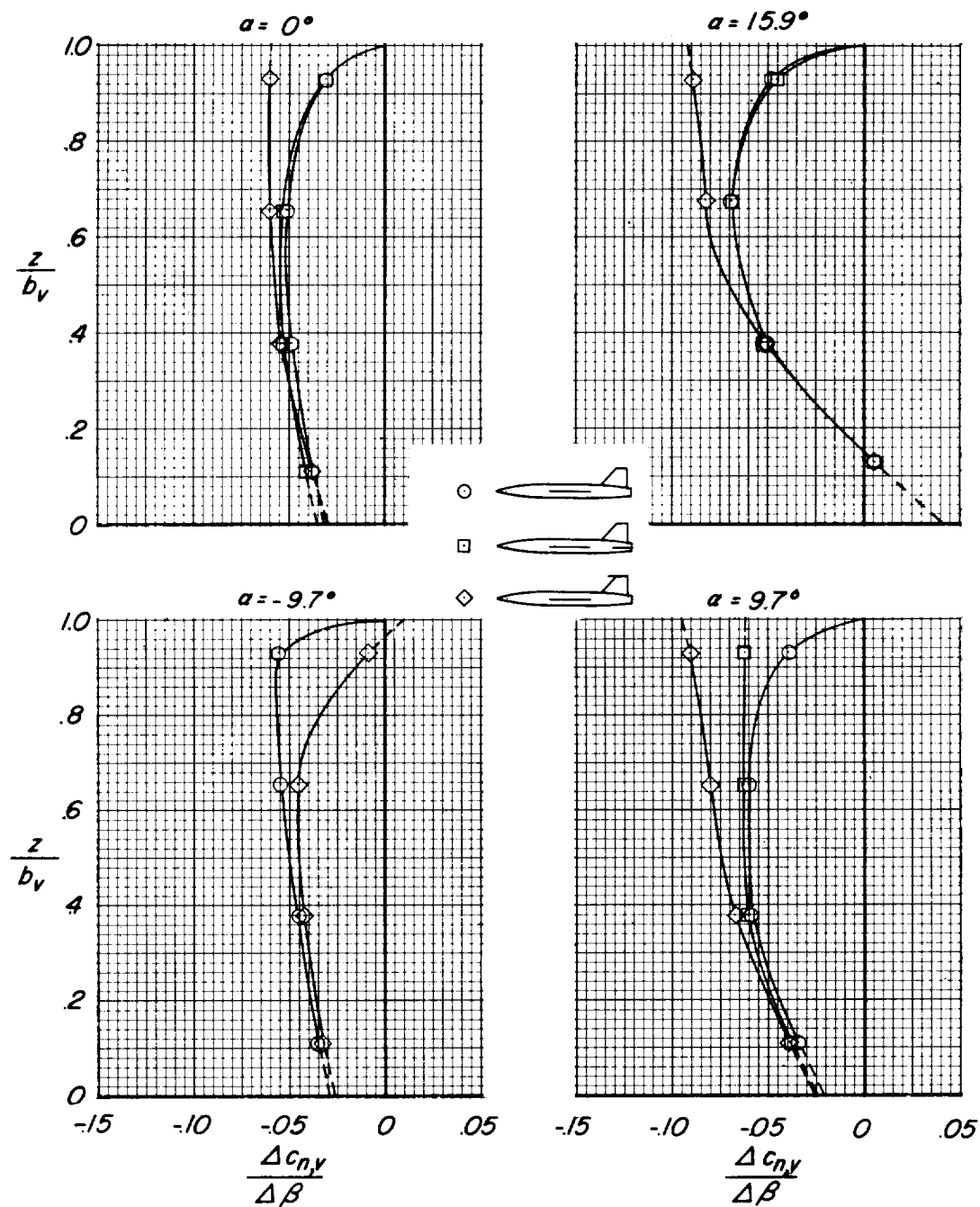
(c) M = 0.85.

Figure 8.- Continued.
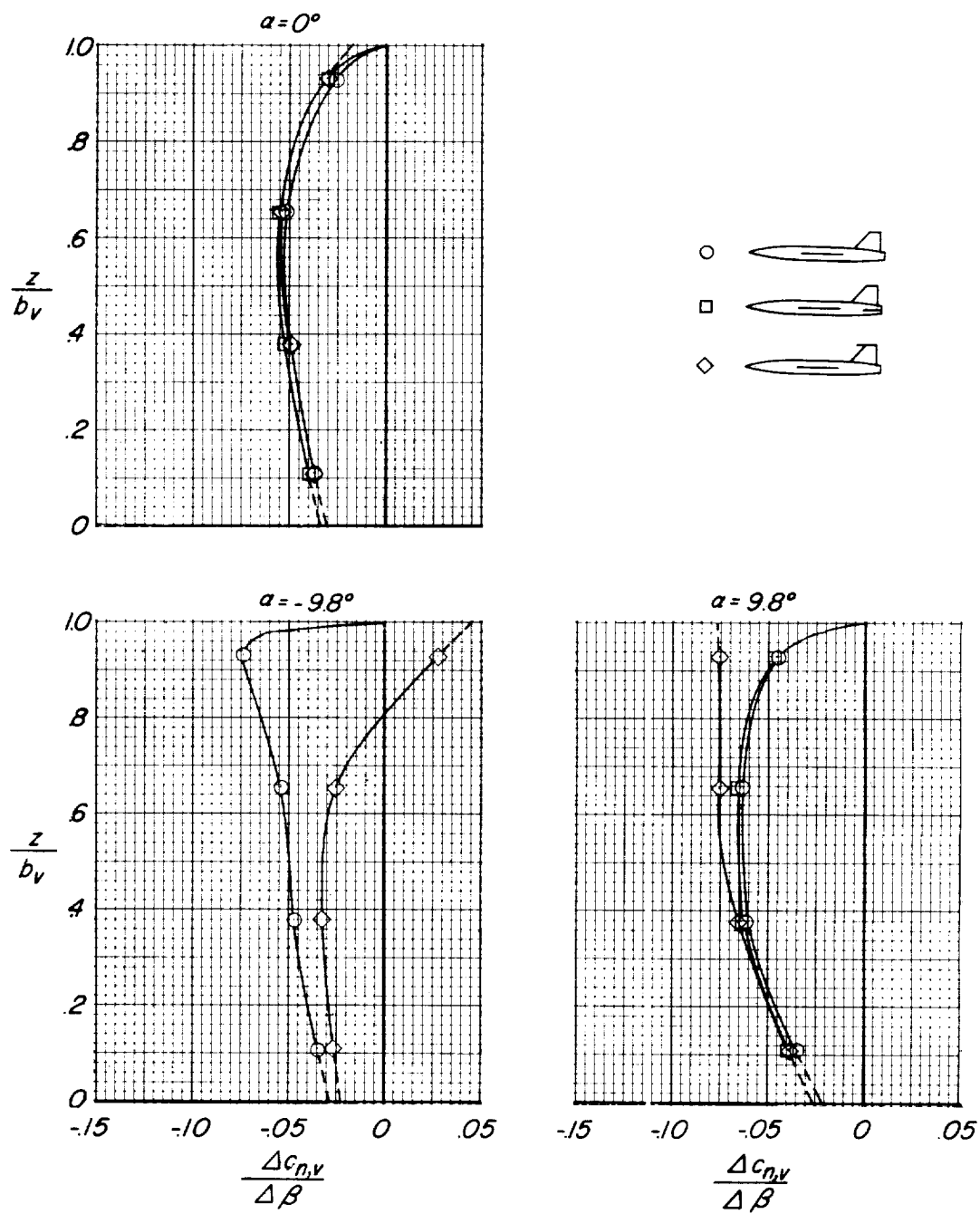
(d) M = 0.90.

Figure 8.- Continued.
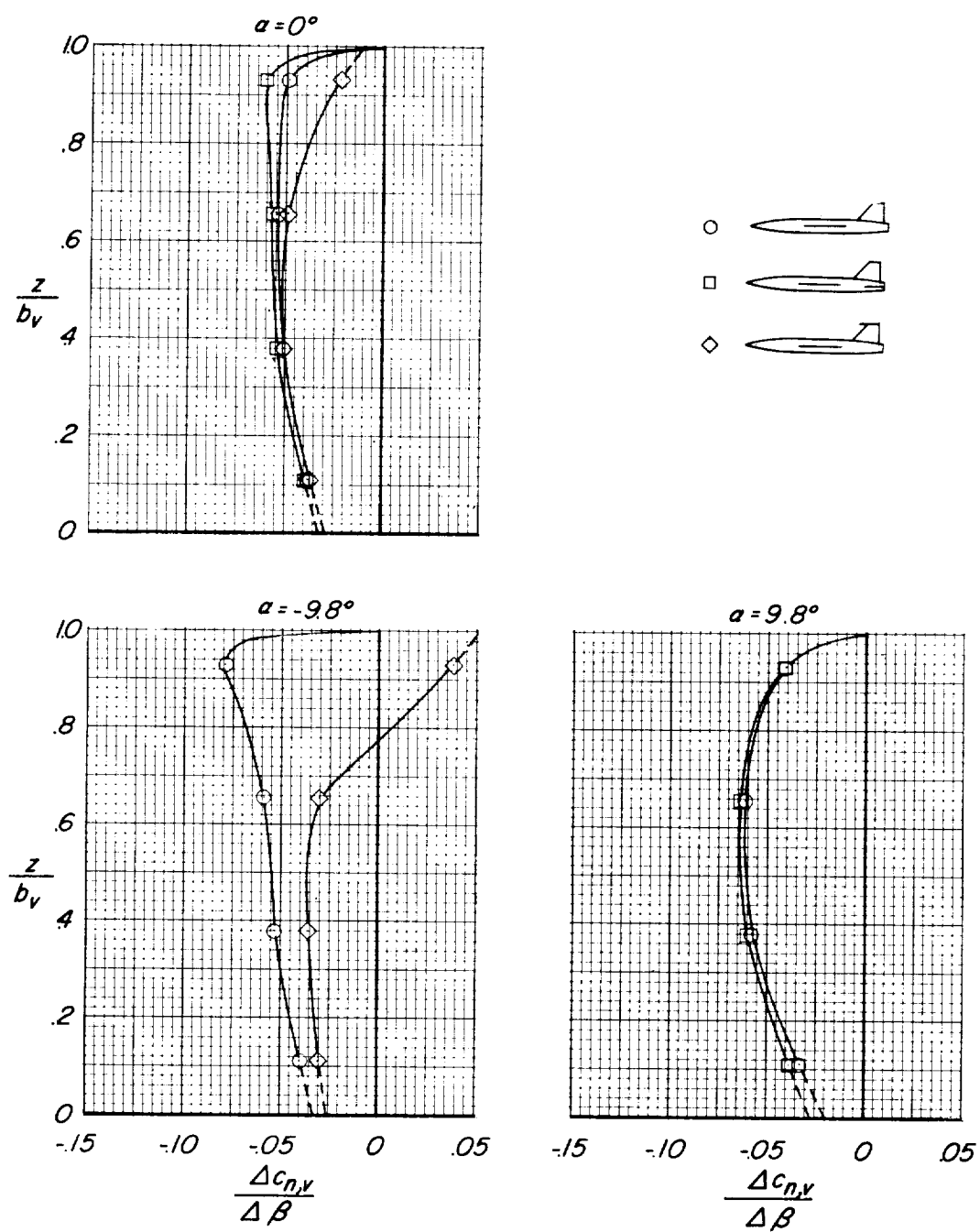
(e)   M = 0.92.

Figure 8.- Concluded.
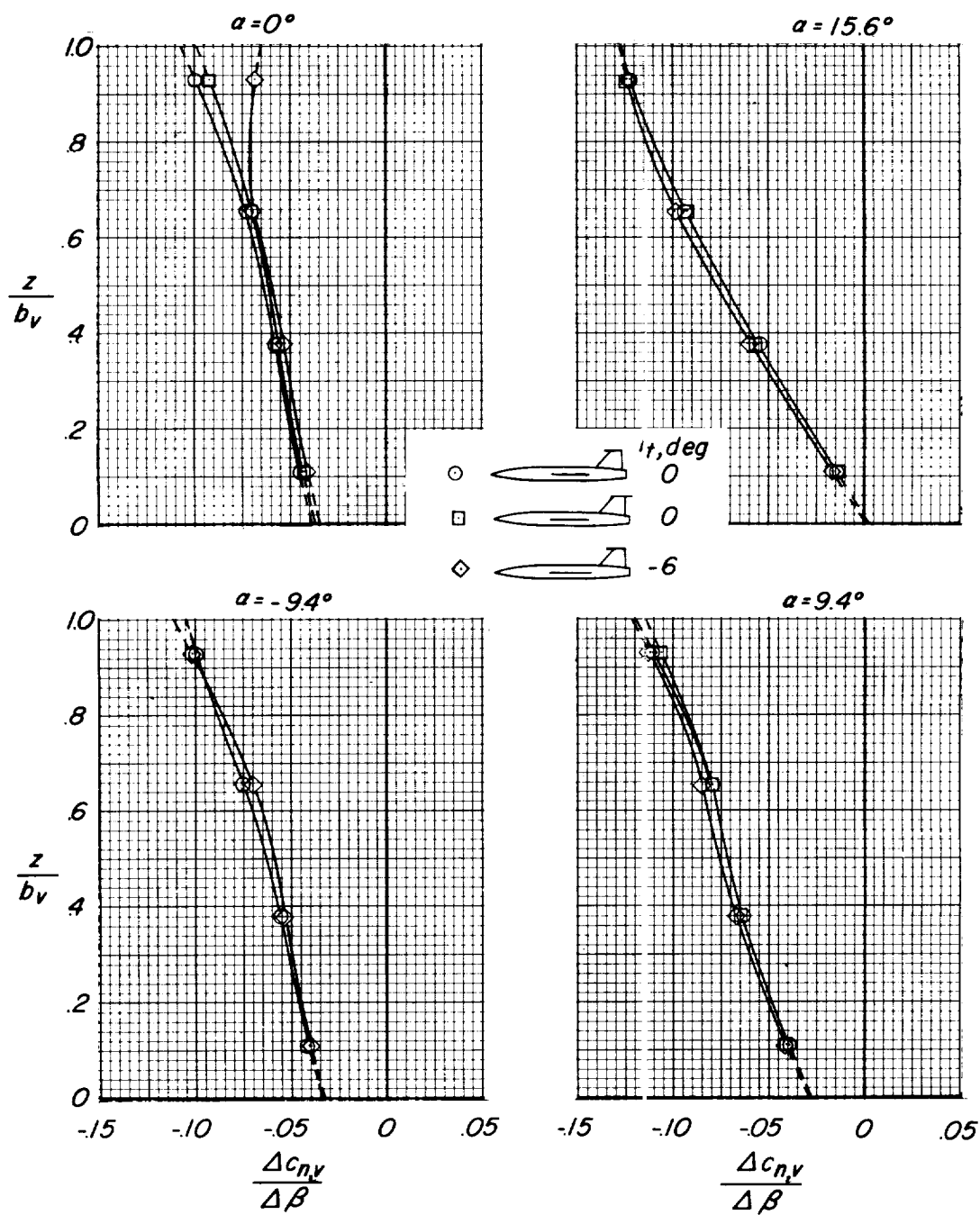
(a) M = 0.60.

Figure 9.- Effect of horizontal-tail position and incidence on the
spanwise variation of vertical-tail section normal-force coefficient
per degree of sideslip. $\Delta\beta \approx 8°$.
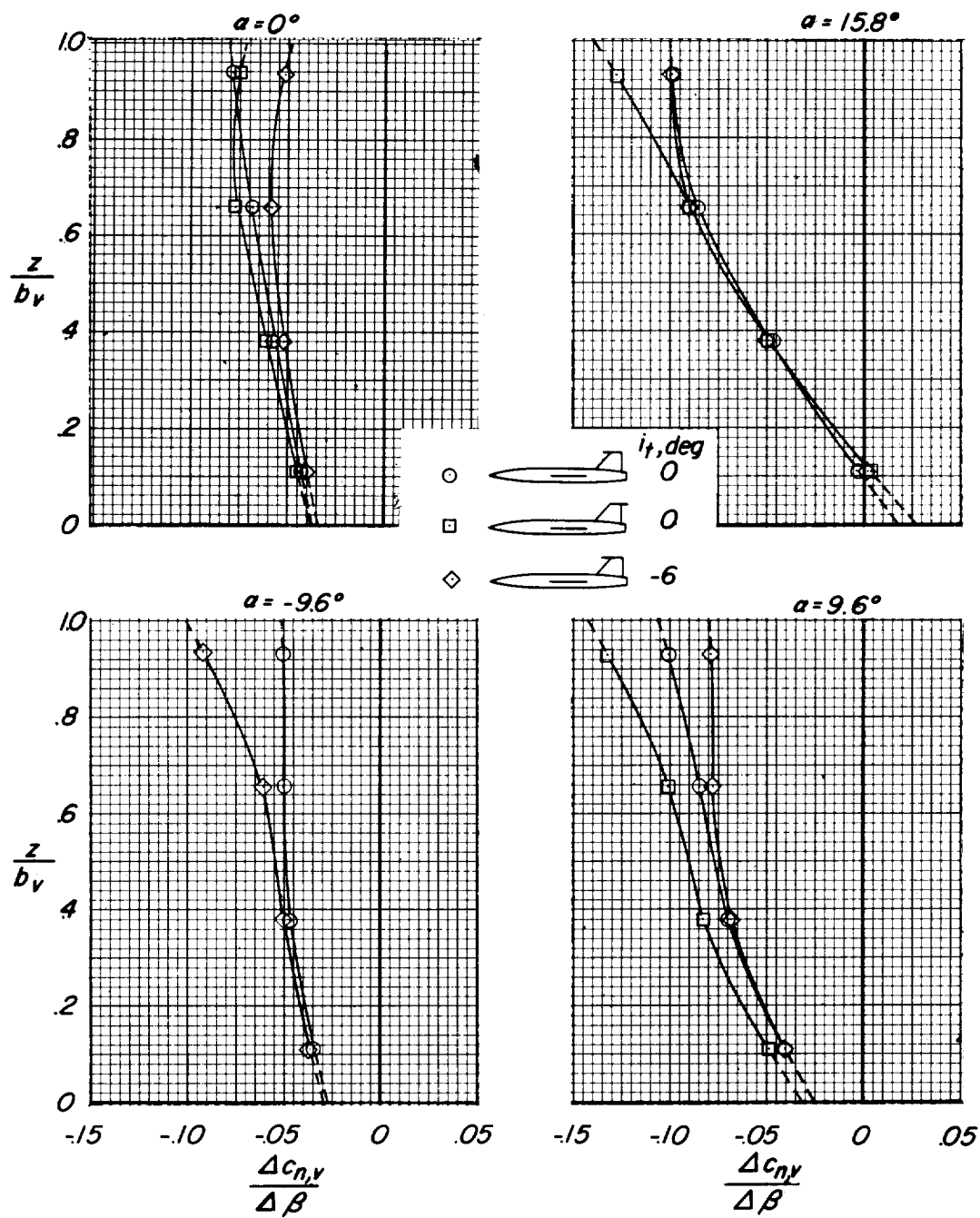
(b)   M = 0.80.

Figure 9.- Continued.
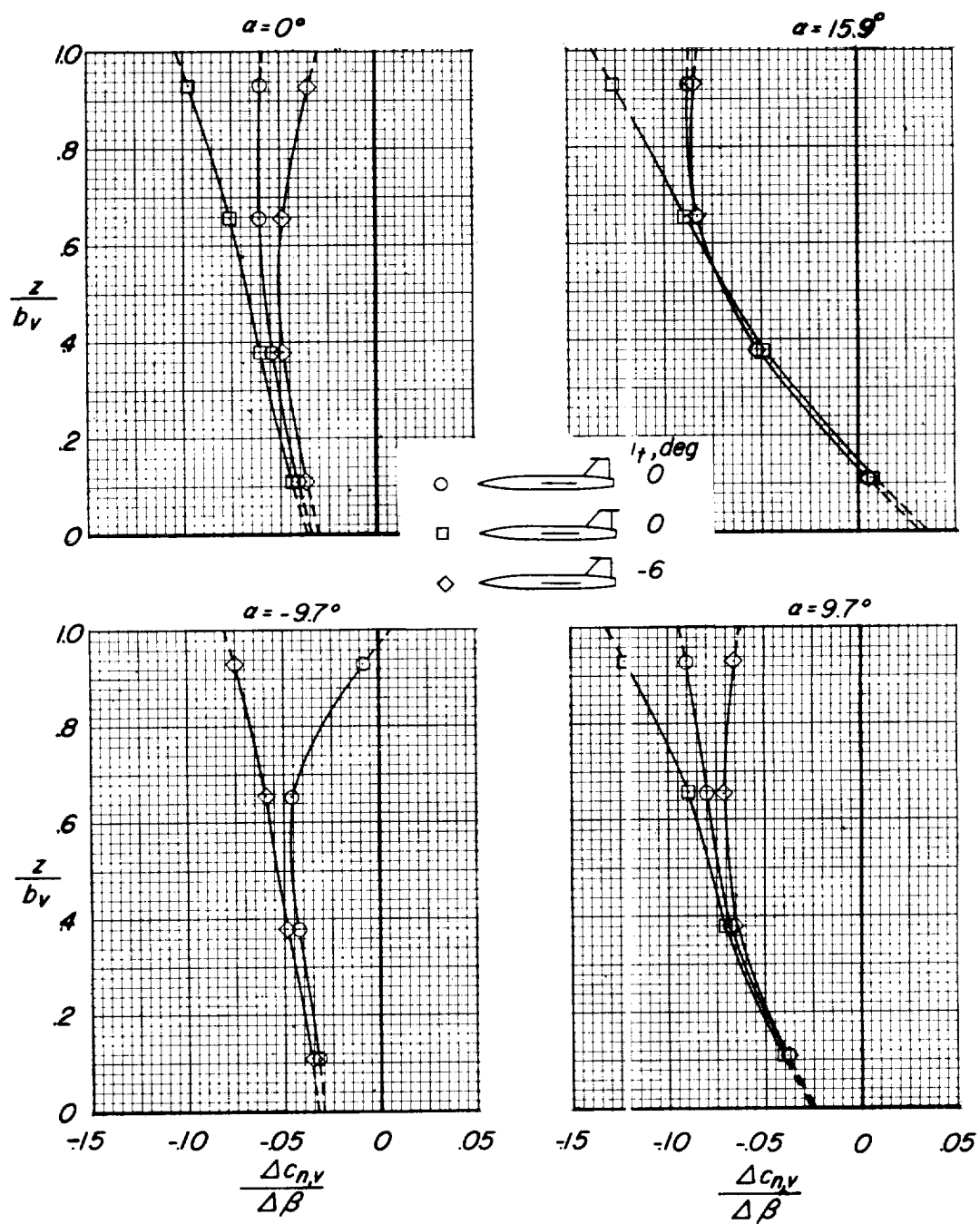
(c)   M = 0.85.

Figure 9.- Continued.
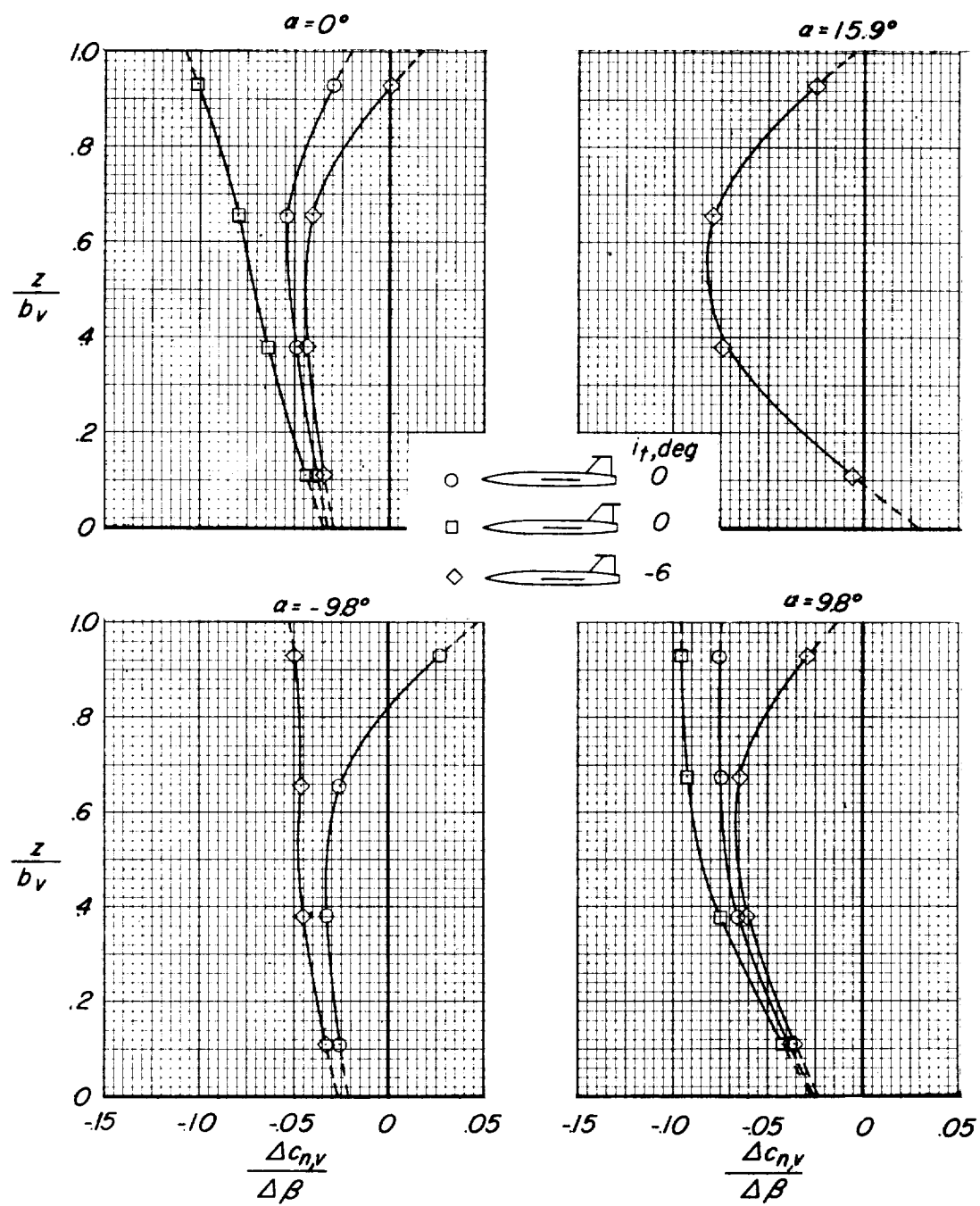
(d) M = 0.90.

Figure 9.- Continued.
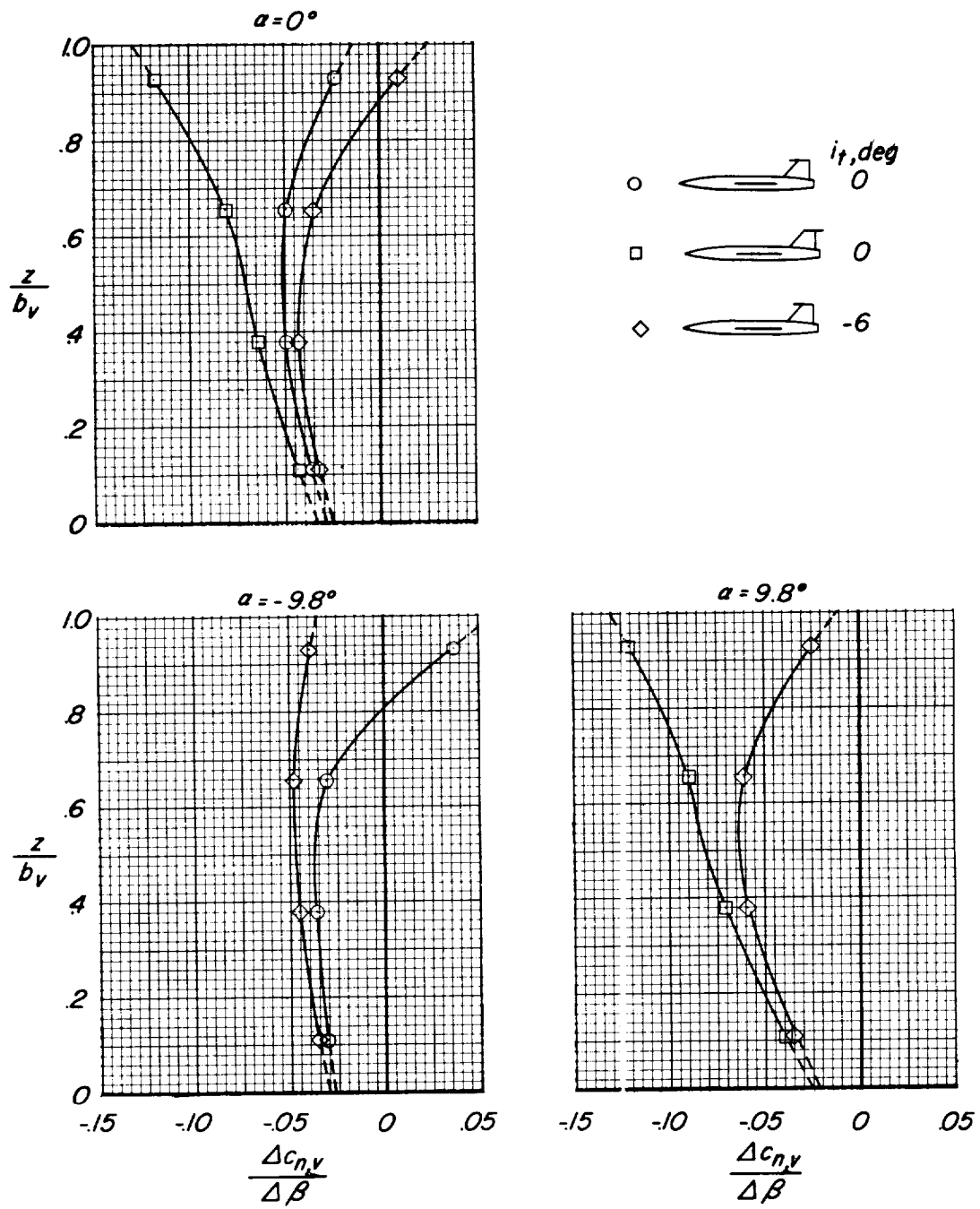
(e) M = 0.92.

Figure 9.- Concluded.

(a) $\alpha = 0^\circ$.

Figure 10.- Effect of horizontal-tail position and incidence on the chordwise location of vertical-tail local centers of pressure. $i_t = 0^\circ$ (except where noted).
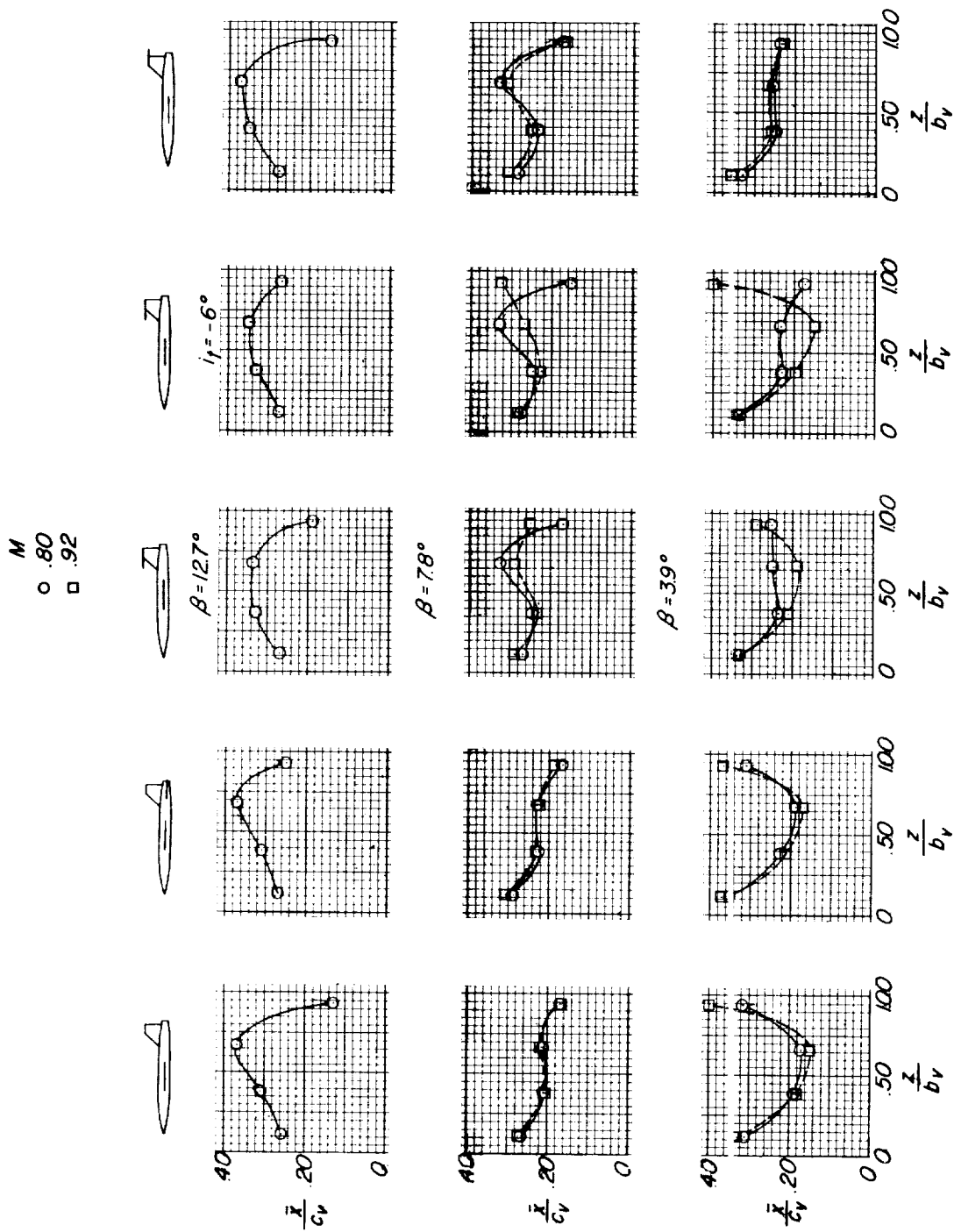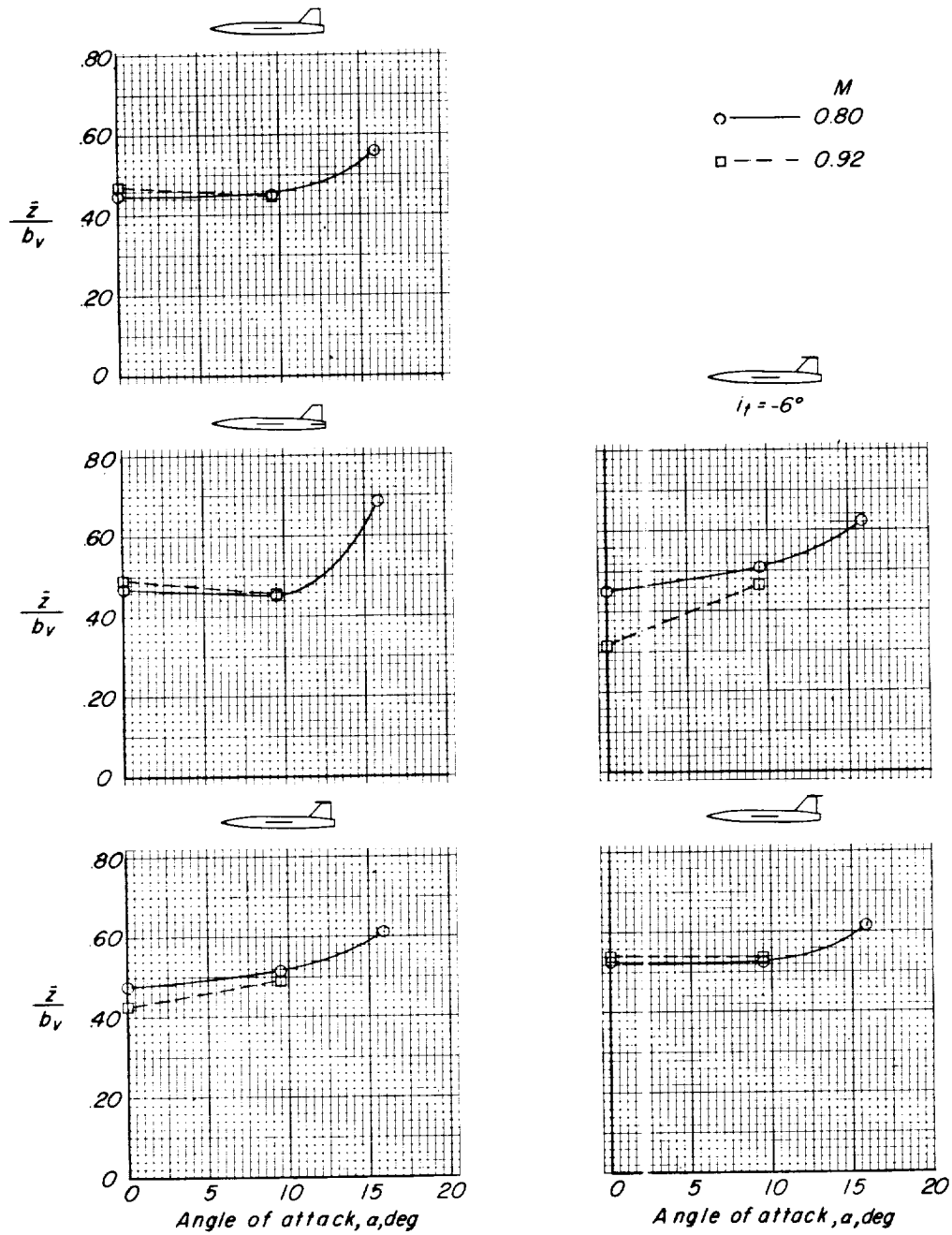
(b) $\alpha \approx 9.6°$.

Figure 10.- Continued.

M
0.80

$i_t = -6°$

$\beta = 12.7°$

$\beta = 7.8°$

$\beta = 3.9°$
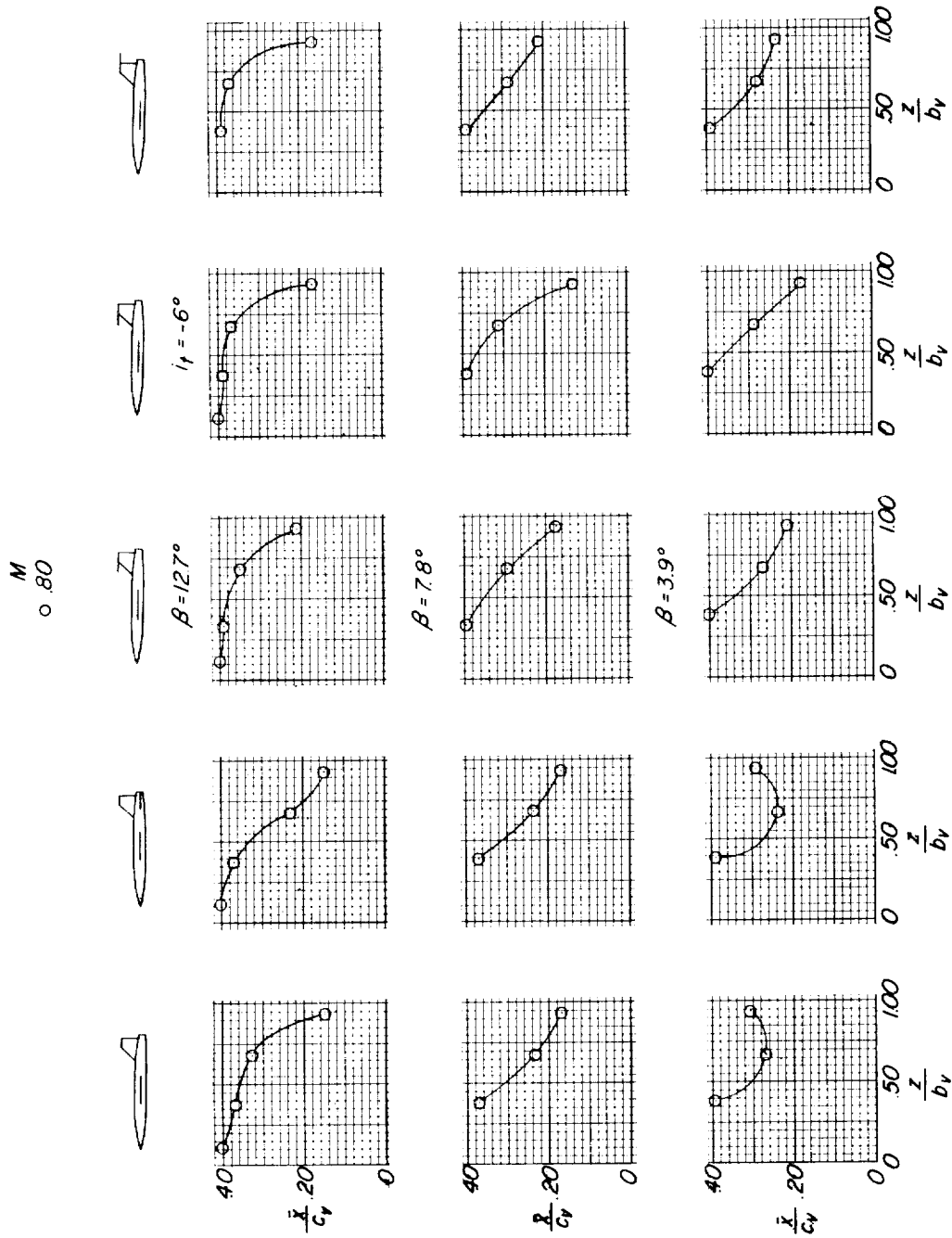
(c) $\alpha \approx 15.3°$.

Figure 10.- Concluded.

$$\frac{\bar{z}}{b_v}$$

M
○——— 0.80
□ — — 0.92

$i_f = -6°$

Angle of attack, a, deg

Angle of attack, a, deg

(a) Spanwise variation.

Figure 11.- Effect of horizontal-tail position and incidence on the variation of the center of pressure of the total vertical-tail load with angle of attack. $\Delta\beta \approx 8°$.

(b) Chordwise variation.

Figure 11.- Concluded.