



Simulated Solar Flare X-Ray and Thermal Cycling Durability Evaluation of Hubble Space Telescope Thermal Control Candidate Replacement Materials

Kim K. de Groh and Bruce A. Banks
Lewis Research Center, Cleveland, Ohio

Edward A. Sechkar and David A. Scheiman
NYMA, Inc., Brook Park, Ohio

Prepared for the
Fourth International Space Conference on Protection of Materials and
Structures from the LEO Space Environment
sponsored by the University of Toronto and the Institute for Aerospace Studies
Toronto, Canada, April 23–24, 1998

National Aeronautics and
Space Administration

Lewis Research Center

Acknowledgments

The authors would like to thank the following for their contributions to this program: Tom Stueber of NYMA Inc., and Alice Jalics of Cleveland State University for tensile tests, Kevin Malinowski of Cleveland State University for assistance with thermal cycling, and Dr. Steven Pepper of NASA Lewis Research Center for x-ray spectrum characterization. We would also like to thank Patti Hansen (the HST MLI FRB Chair) and Jacki Townsend of NASA Goddard Space Flight Center, and Joyce Dever of NASA Lewis Research Center for helpful discussions.

Trade names or manufacturers' names are used in this report for identification only. This usage does not constitute an official endorsement, either expressed or implied, by the National Aeronautics and Space Administration.

Available from

NASA Center for Aerospace Information
7121 Standard Drive
Hanover, MD 21076
Price Code: A03

National Technical Information Service
5285 Port Royal Road
Springfield, VA 22100
Price Code: A03

SIMULATED SOLAR FLARE X-RAY AND THERMAL CYCLING DURABILITY EVALUATION OF HUBBLE SPACE TELESCOPE THERMAL CONTROL CANDIDATE REPLACEMENT MATERIALS

Kim K. de Groh and Bruce A. Banks
National Aeronautics and Space Administration
Lewis Research Center
Cleveland, Ohio 44135

Edward A. Sechkar and David A. Scheiman
NYMA, Inc.
Brook Park, Ohio 44142

During the Hubble Space Telescope (HST) second servicing mission (SM2), astronauts noticed that the multi-layer insulation (MLI) covering the telescope was damaged. Large pieces of the outer layer of MLI (aluminized Teflon[®] fluorinated ethylene propylene (Al-FEP)) were cracked in several locations around the telescope. A piece of curled up Al-FEP was retrieved by the astronauts and was found to be severely embrittled, as witnessed by ground testing. The National Aeronautics and Space Administration (NASA) Goddard Space Flight Center (GSFC) organized a HST MLI Failure Review Board (FRB) to determine the damage mechanism of the Al-FEP in the HST environment, and to recommend a replacement thermal control outer layer material to be installed on HST during the subsequent servicing missions. Candidate thermal control replacement materials were chosen by the FRB and tested for environmental durability under various exposures and durations by GSFC and NASA Lewis Research Center (LeRC). This paper describes durability testing at LeRC of candidate materials which were exposed to charged particle radiation, simulated solar flare x-ray radiation and thermal cycling under load. Samples were evaluated for changes in solar absorptance and tear resistance. Descriptions of environmental exposures and durability evaluations of these materials are presented.

1. Introduction

The Hubble Space Telescope (HST) was launched on April 25, 1990 into low Earth orbit and is the first mission of NASA's Great Observatories program. It is a telescope capable of performing observations in the near-ultraviolet, visible and near-infrared (0.115 to 1000 μm). The HST was designed to be serviced on-orbit to upgrade scientific capabilities. The first servicing mission (SM1) occurred in December 1993, after 3.6 years in

space. The second servicing mission (SM2) was in February 1997, after 6.8 years in space. Servicing missions are planned for mid 2000 and early 2003.¹

The HST is covered with two types of thermal control materials, radiators and multi-layer insulation (MLI) blankets, which passively control temperatures on-orbit. Both of these thermal control materials have metallized-FEP as the exterior layer. During SM1 astronauts retrieved and returned Al-FEP and Ag-FEP MLI blanket material. The Al-FEP (2 blankets retrieved) covered the two magnetometers, and the Ag-FEP MLI covered the solar array drive arm (SADA). Analysis of the retrieved blanket materials revealed that the FEP exterior layer was embrittled on high solar exposure surfaces.^{2,3} Surfaces which received the highest solar exposures (16,670 equivalent sun hours (ESH) for the Al-FEP, and 20,056 ESH for the Ag-FEP) had some small through-thickness cracks in the 5 mil FEP at stress locations.^{2,3} Solar facing bonded 2 mil Al-FEP on the SADA power harness, which was also retrieved, had many cracks and had lost total mechanical integrity (pulverization).⁴

During SM2, severe cracking of the MLI outer layer material was observed on the Light Shield (LS), Forward Shell (FS) and Equipment Bays of the telescope. Astronaut observations combined with photographic documentation of HST taken during SM2 has revealed extensive cracking of the MLI in many locations around the telescope, with solar facing surfaces being particularly damaged.¹ The MLI outer layer at several of the longest cracks was observed to be curled up and lifted away from the next MLI layer. Figure 1 shows severely degraded MLI on HST as photographed during SM2.

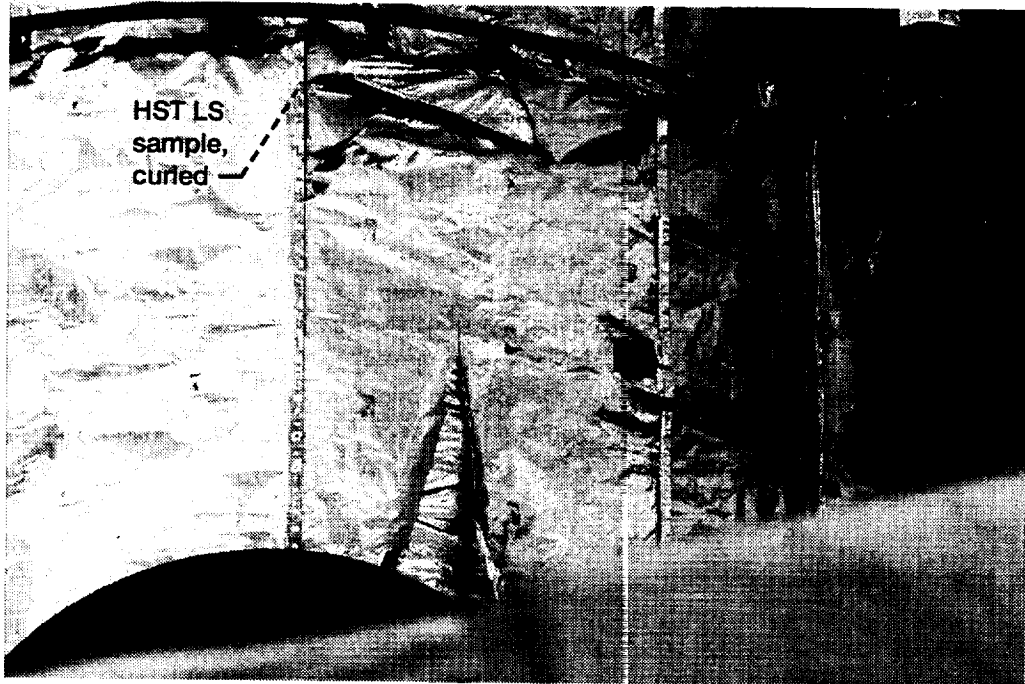


Figure 1. Extensive tearing/curling damage to MLI exterior layer of HST as photographed during SM2. The tightly curled-up piece was cut off by an astronaut and retrieved for analysis (SM2 LS sample) prior to patching the area.

The worst of the MLI outer layer cracks were patched during the last EVA (extravehicular activity) day. Contingency patches were placed over the cracks on the light shield, Bay 8 and Bay 10. Due to limited material, Lanyards were used on Bay 7 to hold the cracked MLI in place. Prior to patching the upper light shield crack, the tightly curled outer layer was removed and retrieved for post-mission analysis (see Figure 1). After SM2, GSFC established a HST MLI Failure Review Board (FRB) to determine the cause of degradation of the FEP MLI on HST, and to identify a replacement material to be installed during subsequent servicing missions. Extensive investigation of the optical, chemical and mechanical properties of the retrieved SM2 FEP have been conducted and compared with SM1 and pristine FEP results.^{1,5,6} Simulated LEO environmental exposure testing of pristine FEP was also conducted to help determine the cause of degradation of FEP on HST.^{7,8}

The replacement thermal control material was required to meet HST's stringent thermal requirements (end-of-life $\alpha_s/\epsilon \leq 0.28$) and to maintain its structural integrity for at least 10 years on-orbit. Candidate replacement materials were chosen and various sets of materials were exposed to combinations of electron/proton radiation, atomic oxygen, simulated solar flare x-rays, thermal cycling and near ultraviolet (NUV) radiation at various facilities in order to evaluate their LEO durability. A summary of all the sets of candidate materials has been written by Townsend et. al.⁹ The simulated solar flare x-ray and thermal cycling exposure testing which was conducted at NASA Lewis Research Center (LeRC) is discussed in this paper.

2. Materials and Experimental Procedures

2.1 CANDIDATE MATERIALS

The HST MLI FRB members suggested and considered seventeen replacement thermal control materials. Each replacement material was scored using a multiplicative evaluation formula based on nine performance criteria chosen by the board.⁹ Six candidate materials were chosen from the list of seventeen for environmental durability testing (materials 1-6). Because of the stringent thermal requirements (end-of-life $\alpha_s/\epsilon \leq 0.28$) replacement choices were limited. Several types of metallized FEP bonded to scrim were chosen because of their excellent optical properties, and because the radiator surfaces on HST (bonded FEP, as opposed to blanket material) maintained their structural integrity as documented during SM2.⁹ Four additional materials were also chosen for testing: the current Al-FEP MLI material (material 8) to be used to verify environmental damage as witnessed on HST, an Optical Coating Laboratory, Inc. (OCLI) coated sample (material 7) because it is used on some HST exterior surfaces and HST management wanted to check its performance, and two materials (9 and 10) chosen by the FRB Chair which were fundamentally different than the first six materials.⁹ These ten materials are listed in Table 1. Materials 3, 4 and 6 (vacuum deposited Al) were purchased from Dunmore, who used their proprietary non-UV-darkening polyester adhesive. Materials 1, 2 and 5 (silver/Inconel) were purchased from Sheldahl, who used their proprietary non-UV-darkening polyester adhesive. Material 7 came from GSFC stock, material 8 was supplied by Lockheed Martin Missiles and Space, material 9 was fabricated at GSFC, and material 10 was provided by AZTechnology.⁹

Table 1. HST Thermal Control Candidate Replacement Materials

Material	Sample Id.	Candidate Material
1	B1.1 and M2.1	10 mil FEP/Ag/Inconel/adhesive/Nomex [®] (polyphenylene isophthalate) scrim
2	B1.2 and M2.2	5 mil FEP/Ag/Inconel/adhesive/fiberglass scrim/adhesive/2 mil Kapton [®]
3	B1.3 and M2.3	10 mil FEP/Al/adhesive/Nomex [®] scrim
4	B1.4 and M2.4	5 mil FEP/Al/adhesive/fiberglass scrim/adhesive/2 mil Kapton [®]
5	B1.5 and M2.5	5 mil FEP/Ag/Inconel/adhesive/Nomex [®] scrim
6	B1.6 and M2.6	5 mil FEP/Al/adhesive/Nomex [®] scrim
7	B1.7 and M2.7	OCLI multi-layer oxide UV blocker/2 mil white Tedlar [®]
8	B1.8 and M2.8	5 mil FEP/Al (current HST material)
9	B1.9 and M2.9	SiO ₂ /Al ₂ O ₃ /Ag/Al ₂ O ₃ /4 mil stainless steel
10	B1.0 and M2.0	Proprietary Teflon [®] FEP/AZ93 White Paint/Kapton [®]

As previously mentioned, sets of samples were subsequently exposed to different environmental exposures (electron/proton radiation, atomic oxygen, simulated solar flare x-rays, thermal cycling and near ultraviolet radiation) at various facilities. The samples that were exposed to simulated solar flare x-rays at LeRC were B1.1-B1.8 and B1.0, and M2.1-M2.8. The samples that were exposed to thermal cycling at LeRC were samples B1.1-B1.8 (samples M2.1-2.8 were thermal cycled at GSFC). Prior to environmental exposure, all samples were purposely cut through approximately one quarter of its width, 5.08 cm (2") from the top.⁹ This was done to provide a stress concentration area during environmental testing. Samples in the "B" set were exposed to charged particle radiation at The Boeing Company's Radiation Effects Laboratory (Boeing). These samples were 3.81 cm (1.5") wide and 12.7 cm (5") long. Samples in the "M" set were exposed to charged particle radiation at the NASA Marshall Space Flight Center (MSFC) Space Environmental Effects (SEE) facility. These 3.81 cm (2") wide samples were cut to 6.7 cm (2.6") in length to fit in MSFC's facility.

2.2 OPTICAL PROPERTY CHARACTERIZATION

Optical properties were obtained using a Perkin-Elmer λ -9 spectrophotometer equipped with a 150 mm integrating sphere. Total spectral reflectance and total spectral transmittance were obtained from 250 to 2,500 nm. The spectral data were convoluted into the air mass zero solar spectrum over the same wavelength range and integrated to obtain solar total reflectance (ρ_t) and solar total transmittance (τ_t). Solar absorptance (α_s) was calculated by subtracting ρ_t from 1 for the opaque samples, and by subtracting $\rho_t + \tau_t$ from 1 for the transmitting samples. Instrument repeatability is ± 0.005 . The thermal emittance (ϵ) of Al-FEP retrieved during SMI did not change with environmental exposure,² and therefore was not measured in this test program. Typically, the ϵ for 5 mil Al-FEP is 0.81.¹⁰

2.3 SURFACE AND MASS CHARACTERIZATION

Mass measurements were obtained using a Sartorius balance. Three readings were obtained and averaged for each measurement. The balance has an uncertainty of $\pm 50 \mu\text{g}$. Overall photographs of the samples were obtained using a Polaroid Land camera. Optical micrographs and sample examinations were conducted using an Olympus SZH stereo microscope. Micrographs were obtained at magnifications between 12 and 101 times. A JEOL 6100 scanning electron microscope (SEM) was used for high magnification imaging. Samples were coated with a thin ($\approx 75 \text{ \AA}$) palladium conductive coating prior to SEM examination.

2.4 CHARGED PARTICLE RADIATION

Samples were exposed to charged particle radiation at either MSFC or at Boeing prior to environmental exposure at LeRC. At MSFC, the M samples were exposed to 10 years equivalent HST dose (SM3-2010) of electron and proton radiation.¹¹ Samples were exposed to 50 keV, 220 keV (for 10 mil FEP) or 500 keV (for 5 mil FEP) electrons and 700 keV protons simulating the dose versus depth profile which was calculated for HST. The samples (2 inch wide) were loaded under a 10 lb. tensile load during exposure. Samples were desired to be stressed during environmental exposures. A 10 lb. load was applied to the samples because that is the load (for a 2 inch wide sample) which approximates the stress calculated to be necessary to hold 5 mil FEP flat (on HST) when it would otherwise curl-up tight like the retrieved SM2 sample when unconstrained. At Boeing, the B samples were exposed to a 10 year fluence of electron and proton radiation.¹² Rather than simulating dose versus depth, Boeing simulated the total fluence using 40 keV electrons ($3.1 \times 10^{13} \text{ e/cm}^2$) and 40 keV protons ($2.7 \times 10^{10} \text{ p/cm}^2$). Ultraviolet radiation from a xenon arc source (i.e. a continuum from 200-400 nm) illuminated the samples at approximately 1.2 UV suns intensity during radiation exposure, resulting in approximately 5 UV solar hours exposure. These samples (1.5 inch wide) were also loaded under a 10 lb. tensile load during irradiation.

2.5 SIMULATED SOLAR FLARE X-RAY EXPOSURE

The x-ray fluences predicted for HST from SM3 to the year 2010 are 397.4 J/m^2 for 1-8 \AA x-rays, 28.05 J/m^2 for 0.5-4 \AA x-rays and 0.00967 J/m^2 for 0.124-0.5 \AA x-rays. These fluence values were calculated at GSFC based on GOES data, which were extrapolated, based on the 11 year solar cycle. The fluence is substantially higher for the 1-8 \AA x-rays (12,396-1,550 eV) than for the more energetic x-rays. These lower energy solar flare x-rays are often referred to as soft x-rays. Although FEP is transparent at visible wavelengths, it is highly absorbing between the vacuum ultraviolet to soft x-ray range of 25 to 3000 eV.⁸ If one looks at the x-ray attenuation length versus photon energy for FEP, as seen in Figure 2, it becomes apparent that energy below 3000 eV is primarily absorbed in the top 20 μm of the film. Energy between 3000 eV and 6000 eV is primarily absorbed in the bulk of a FEP film 127 μm thick. The transmittance versus energy curve for 127 μm thick FEP shows that 84 percent of photons with energy of 10,000 eV pass through the

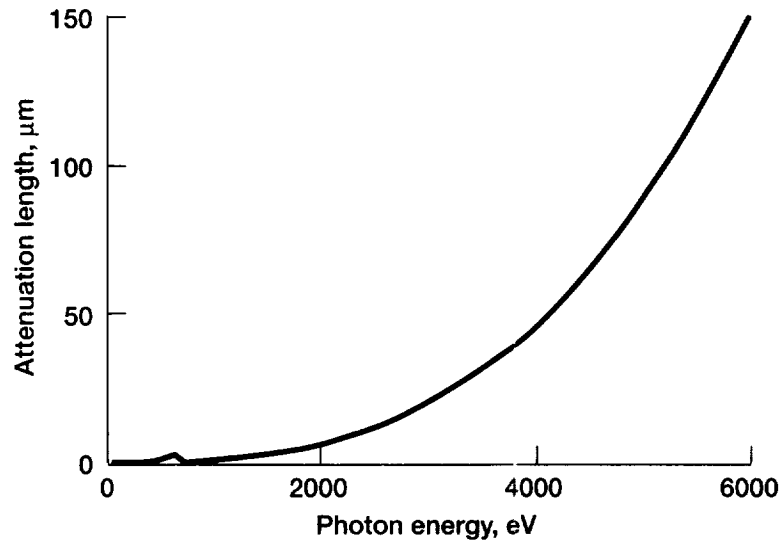


Figure 2. X-ray attenuation length versus photon energy (0-6000 eV) for FEP.

film.⁸ It was therefore concluded that irradiation with continuous x-rays below 10 keV would provide the x-ray energies that should be necessary for bulk film degradation as seen on HST.

The samples were exposed to simulated solar flare x-rays in a modified electron beam evaporator system. A water-cooled molybdenum (Mo) target was irradiated with a 10 keV electron beam. The target was angled to allow the highest flux of x-rays to irradiate the candidate materials. The electron beam current was run low enough to prevent any evaporation of the target material. A model AXUV-20HEI absolute XUV silicone photodiode, produced by International Radiation Detectors Inc., was used to measure the x-ray flux during each sample exposure run. The photodiode has 100 percent quantum efficiency over the range of photon energies produced by the source. Two sheets of 2 μm Al foil were used as a barrier between the target and the photodiode and samples during the exposures. The Al foil blocked energetic electrons from the target, and blocked the detector from visible light from the electron beam emitter. A photographic cloth was used to block room light to the detector.

To maximize the x-ray flux, only one candidate sample was exposed at a time. With each candidate sample, a witness Al-FEP (5 mil FEP) tensile sample was exposed to evaluate the effect of x-ray radiation on the tensile properties of Al-FEP. Figure 3 is a drawing of the electron beam system configured to expose the HST candidate samples to x-rays. Figure 4 shows the x-ray spectrum as a function of energy for a 10 keV electron beam bombarding a Mo target, and with the x-rays passing through a 4 μm Al barrier. This spectrum was obtained using a windowless energy-dispersive spectrometry (EDS) detector on a scanning electron microscope. With an electron beam energy of 10 keV, a Mo target will not produce any K-alpha characteristic radiation (17,476 eV), eliminating the very intense K-alpha characteristic peak. Yet, as can be seen in Figure 4, there is still a fairly substantial Mo L-alpha line at 2,293 eV (which appears wider in this

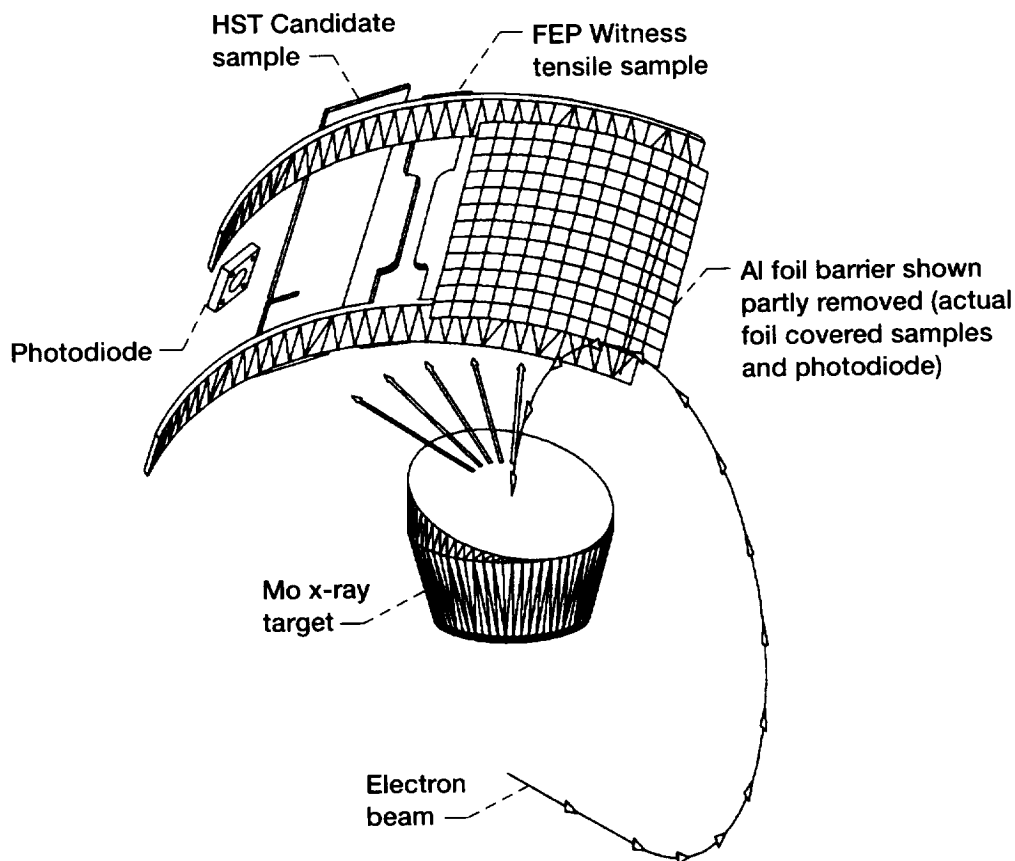


Figure 3. Drawing of the electron beam system configured to expose the HST candidate samples to Mo x-rays.

spectra than the actual peak width, which is a detector artifact). As can be seen in Figure 4, there is Bremsstrahlung, or continuous radiation between the energy range of 3000 to 6000 eV, and above, where bulk damage is anticipated to occur in FEP. The actual fluence exposures for each candidate sample are listed in Table 2. Due to electron beam gun controller fluctuations, the final fluence values varied somewhat from the desired 397.4 J/m^2 ($457 \pm 50 \text{ \AA}$). After simulated solar flare x-ray exposure characterization, the M samples were returned to GSFC for thermal cycling.

2.6 THERMAL CYCLING

Samples B1.1-B1.8 were thermal cycled in LeRC's Rapid Thermal Cycling Facility. The facility consists of two smaller chambers, hot over cold held at fixed temperatures, and frames that shuttle the samples between the two chambers. The hot side is heated via resistive heater elements and a low flow circulating fan to minimize the gradient. The cold side is cooled using liquid nitrogen which is fed into the chamber through a vented pipe;

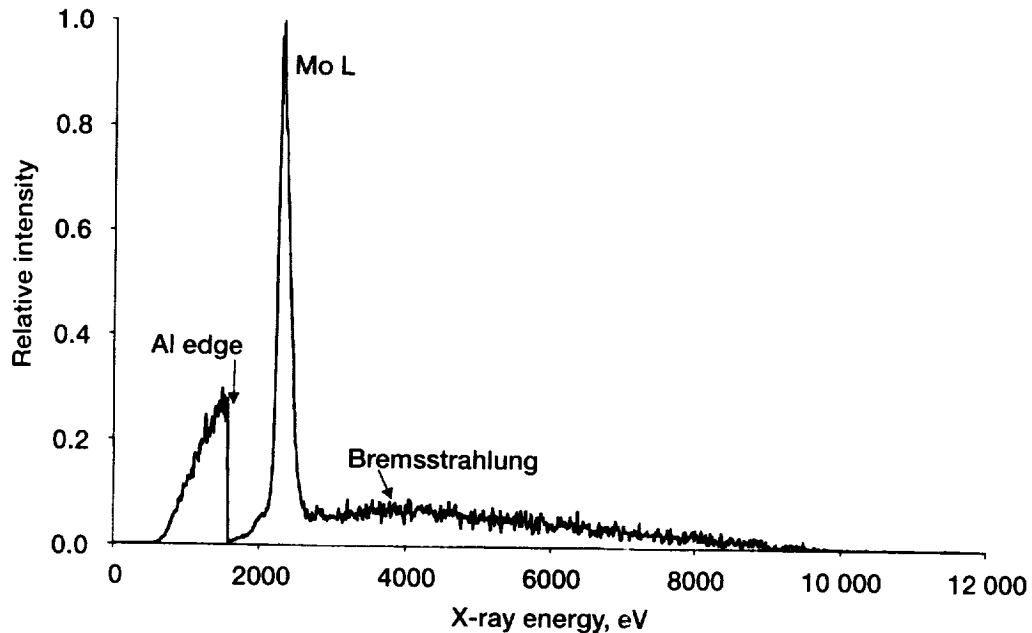


Figure 4. X-ray spectrum as a function of energy for a 10 keV electron beam impacting a Mo target, and with the x-rays passing through a 4 μm Al barrier.

the nitrogen sprays away from the samples. Gaseous nitrogen fills the chambers and provides an inert atmosphere. Additionally, there is a nitrogen line to provide cooling on the hot side and a heater on the cold side to improve the temperature stability, and to provide some temperature profiling if needed. Typically, samples are mounted to frames attached to lever arms. The lever arms are air driven which raises and lowers the samples between the two chambers. A flange on the end of the frame seals the frame in each chamber preventing thermal leakage between the two chambers. The facility is computer controlled; all valves and relays are operated based on temperature sensors in the chambers and on or near the samples. This control allows the samples to cycle as rapidly as possible.

The candidate samples were cycled from -100°C to $+50^{\circ}\text{C}$. The thermal mass of the samples was extremely low and therefore a thermocouple was not attached to the samples but only placed freely next to them. The candidate samples were each tensioned in metal fixtures under 10 lb. loads, (1,778 psi for sample B1.8) and four of these fixtures were then mounted to each of the two chamber frames. The weight of these chamber frames was above the capability of the air cylinders and counterweights were added to the lever arms to obtain transition between the two temperature chambers. The chambers are set up vertically and therefore must work with or against gravity. The additional dynamic forces caused by the motion of the frames during transition may have added some non-thermal stresses to the coupons during cycling. These samples were cycled for 1000 cycles.

Table 2. X-Ray Fluences and Tensile Properties for Witness Al-FEP

Sample	X-Ray Fluence (J/m ²)	Tensile Properties of Companion 5 mil Al-FEP Witness Samples	
		UTS (MPa)	% Elongation
B1.1	431	15.4	163
B1.2	466	15.2	154
B1.3	447	19.1	226
B1.4	541	14.9	151
B1.5	521	16.3	178
B1.6	439	17.3	201
B1.7	488	15.1	160
B1.8	540	14.8	150
B1.0	401	16.9	183
M2.1	411	16.6	190
M2.2	399	14.8	150
M2.3	429	18.3	217
M2.4	414	16.0	177
M2.5	463	14.8	154
M2.6	399	17.4	202
M2.8	515	15.4	161
Average	457 ± 50	16.1 ± 1.3	176 ± 25
5 mil FEP/Al	0	19.2 ± 1.8	197 ± 20

3. Results and Discussion

3.1 CHARGED PARTICLE RADIATION

Optical micrographs, overall photographs, mass measurements and optical properties of the candidate materials after electron/proton radiation exposure by either Boeing (B samples) or MSFC (M samples) were obtained at LeRC prior to simulated solar flare x-ray exposure. Details of the effect of electron/proton radiation exposure on candidate materials are reported in references 11 and 12. In summary, the MSFC samples experienced small changes in solar absorptance (−0.001 to +0.004) and generally slight mass losses, attributed to vacuum dehydration.¹² The Boeing samples experienced small changes in solar absorptance (−0.003 to +0.006) with radiation exposure.¹¹ The error of these solar absorptance measurements is 0.02, indicating that no change in solar absorptance was measured after electron/proton exposure. Mass measurements were not reported.

3.2 SIMULATED SOLAR FLARE X-RAY EXPOSURE (SAMPLES M AND B)

There were generally very small decreases in mass with x-ray exposure (-0.003 to -0.426%). The mass measurements are provided in Table 3. The optical properties remained essentially unchanged with x-ray exposure (a pristine NIST second surface mirror was optically characterized as a control). The optical properties are listed in Table 4. Optical microscopy showed no propagation, or change in the purposely induced cut mark, or in the general appearance of the samples with x-ray exposure.

There were small decreases in the tensile properties of the companion Al-FEP witness samples exposed to x-ray radiation. The ultimate tensile strength (UTS) decreased from 19.2 ± 1.8 to 16.1 ± 1.3 MPa (2792 ± 263 to 2349 ± 194 psi), and the percent elongation to failure decreased from 197 ± 20 to 176 ± 25 percent for pristine and x-ray exposed samples, respectively. The tensile properties of the witness samples are listed in Table 2. The average x-ray fluence was 457 ± 50 J/m². It has been suggested that solar flare x-rays are primarily responsible for the degradation of FEP on HST.¹³ The x-ray fluence (1-8 Å) for HST at the time of SMI has been calculated by GSFC to be 131.8 J/m² for the magnetometer surfaces which received 11,339 ESH exposure. Yet, the UTS for this sample was 13.6 MPa (1974 psi) and the percent elongation to failure was only 41.0 percent.⁸ These results, along with additional x-ray exposure tests reported by Banks, et. al.,⁸ and vacuum ultraviolet radiation and soft x-ray exposure tests reported by Dever,⁷ indicate that solar flare x-rays alone can not be the primary contributor to degradation of FEP on HST. This is consistent with the fact that MLI degradation occurred on all sides of HST, not only solar facing sides.

Table 3. HST Candidate Samples Mass Measurements

Sample	Post e/p* Mass (g)	Post X-Ray Mass	% Change	Post T.C. Mass	% Change from Post X-ray	% Change from Post e/p*
B1.1	2.75763 ± 0.00010	2.75639 ± 0.00001	-0.045	2.75607 ± 0.00022	-0.012	-0.057
B1.2	1.95599 ± 0.00016	1.95516 ± 0.00004	-0.042	1.95445 ± 0.00004	-0.079	-0.079
B1.3	2.73842 ± 0.00039	2.73756 ± 0.00010	-0.031	2.73648 ± 0.00010	-0.039	-0.071
B1.4	1.93944 ± 0.00019	1.93884 ± 0.00008	-0.031	1.93713 ± 0.00003	-0.088	-0.119
B1.5	1.43776 ± 0.00009	1.43781 ± 0.00002	+0.003	1.43727 ± 0.00003	-0.038	-0.034
B1.6	1.43027 ± 0.00015	1.42919 ± 0.00003	-0.076	1.42852 ± 0.00004	-0.047	-0.122
B1.7	0.48096 ± 0.00006	0.48087 ± 0.00002	-0.019	0.48066 ± 0.00001	-0.044	-0.062
B1.8	1.29512 ± 0.00005	1.29487 ± 0.00008	-0.019	1.29569 ± 0.00004	+0.063	+0.044
B1.0	1.45386 ± 0.00005	1.44767 ± 0.00017	-0.426	-	-	-
M2.1	1.96282 ± 0.00002	1.96165 ± 0.00003	-0.060	-	-	-
M2.2	1.34966 ± 0.00007	1.34761 ± 0.00002	-0.152	-	-	-
M2.3	1.96205 ± 0.00001	1.96047 ± 0.00012	-0.081	-	-	-
M2.4	1.32761 ± 0.00001	1.32544 ± 0.00006	-0.163	-	-	-
M2.5	0.96760 ± 0.00010	0.96634 ± 0.00004	-0.130	-	-	-
M2.6	1.00858 ± 0	1.00706 ± 0.00016	-0.151	-	-	-
M2.8	0.88820 ± 0.00005	0.88817 ± 0.00003	-0.003	-	-	-

Table 4. HST Candidate Samples Optical Properties

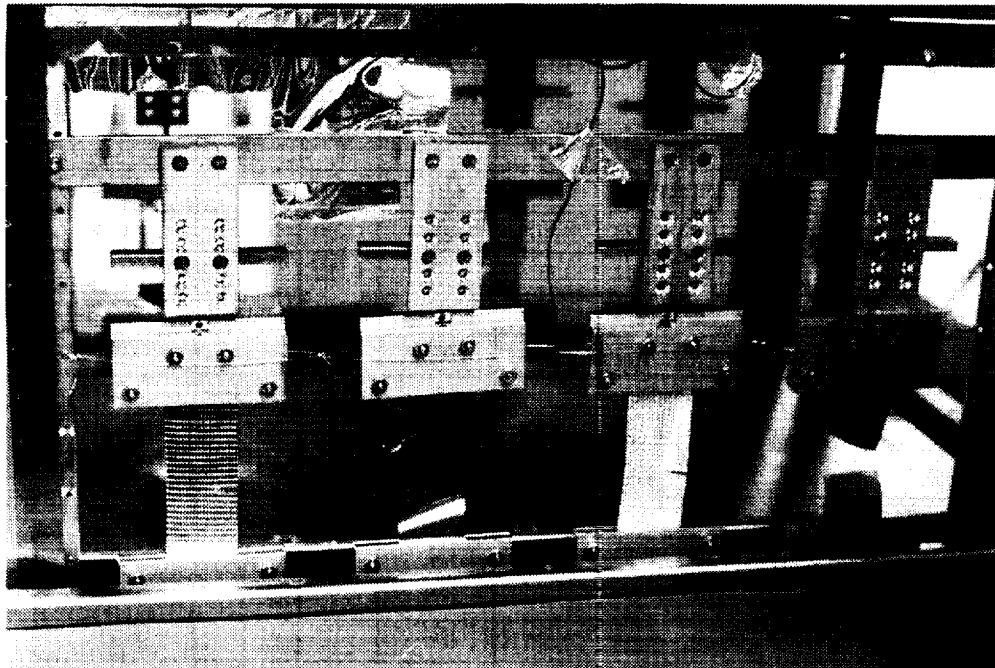
Sample	Post e ⁻ /p ⁺		Post X-Ray		Absolute Change	Post Thermal Cycling		Change from Post X-ray
	ρ_t	α_s	ρ_t	α_s		ρ_t	α_s	
B1.1	0.923	0.077	0.923	0.077	0	0.860	0.140	+0.063
B1.2	0.932	0.068	0.933	0.067	-0.001	0.930	0.070	+0.003
B1.3	0.847	0.153	0.847	0.153	0	0.852	0.148	-0.005
B1.4	0.814	0.186	0.814	0.186	0	0.812	0.188	+0.002
B1.5	0.927	0.073	0.924	0.076	+0.003	0.883	0.117	+0.041
B1.6	0.858	0.142	0.858	0.142	0	0.865	0.135	-0.007
B1.7	0.662	0.338	0.666	0.334 (*0.252)	-0.004	0.667	0.333 (*0.251)	-0.001
B1.8	0.889	0.111	0.888	0.112	+0.001	0.880	0.120	+0.008
NIST 2023	0.883	0.117	0.883	0.117	0	0.883	0.117	0
B1.0	0.844	0.156	0.847	0.153	-0.003	-	-	-
M2.1	0.923	0.077	0.925	0.075	-0.002	-	-	-
M2.2	0.933	0.067	0.936	0.064	-0.003	-	-	-
M2.3	0.838	0.162	0.840	0.160	-0.002	-	-	-
M2.4	0.824	0.176	0.827	0.173	-0.003	-	-	-
M2.5	0.924	0.076	0.927	0.073	-0.003	-	-	-
M2.6	0.859	0.141	0.861	0.139	-0.002	-	-	-
M2.8	0.863	0.137	0.865	0.135	-0.002	-	-	-
NIST 2023	0.883	0.117	0.886	0.114	-0.003			

* Sample has a transmittance of 0.064, which was not measured post e⁻/p⁺ exposure

NIST 2023: second surface Al mirror

3.3 THERMAL CYCLING (SAMPLES B1.1-B1.8)

Samples were originally loaded under 10 lbs. of tension. After 252 cycles for samples B1.1 through B1.4, and 285 cycles for samples B1.5 - B1.8, thermal cycling automatically stopped and the chamber was opened up. Two samples had torn in half (B1.2 and B1.4, both with fiberglass scrim and Kapton[®] backed). Sample B1.8 (the current material, Al-FEP with no scrim) had torn about 90% through the width of the sample. The frames were then inverted to keep the samples from falling over during additional cycling, resulting in a 9 lb. tensile load for the remainder of cycling. Additional thermal cycling did not appear to cause further crack propagation. Figure 5 shows the candidate samples in the test facility after 1000 thermal cycles. The final tension was determined for the intact samples, and ranged between 5.05 lbs. (sample B1.8) to 8.19 lbs. (sample B1.7). Figures 6-8 show several samples as received at LeRC (after electron/proton exposure) and after thermal cycling (and x-ray exposure).

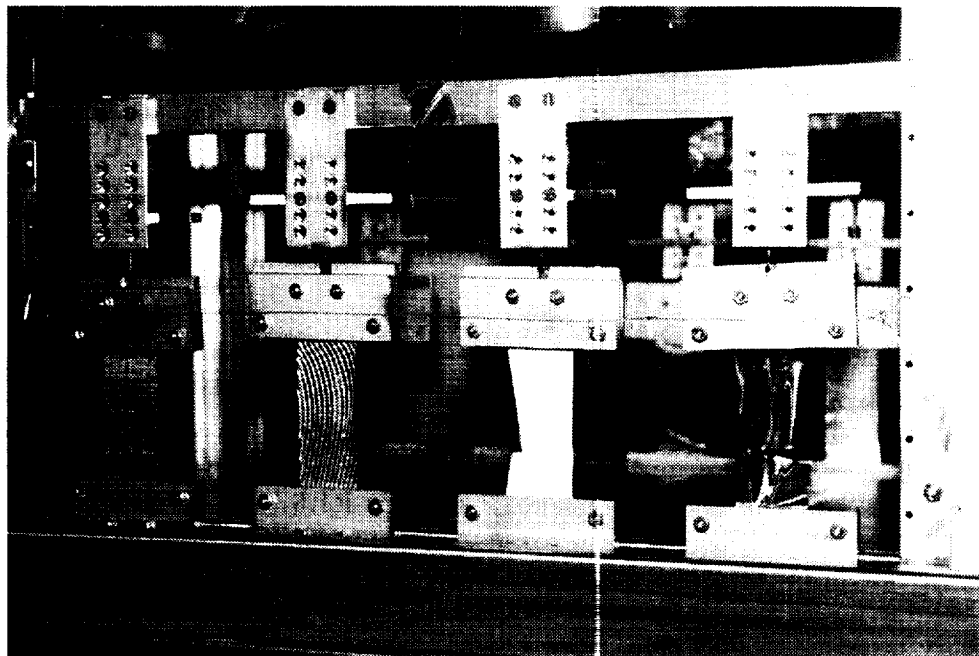


B1.1

B1.2

B1.3

B1.4



B1.5

B1.6

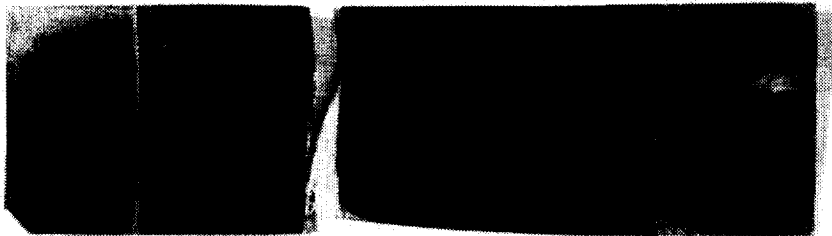
B1.7

B1.8

Figure 5. Candidate samples (B1.1 - B1.8) under tension in the Rapid Thermal Cycling Facility after 1000 thermal cycles.



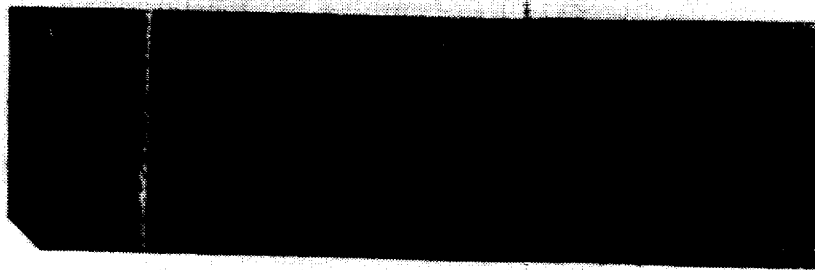
(a)



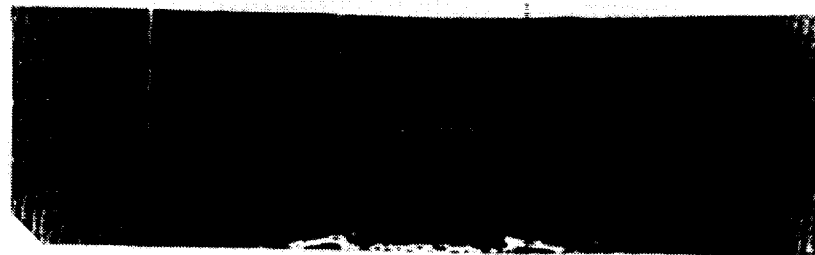
(b)



Figure 6. Candidate sample B1.4 (5 mil FEP/Al/fiber glass scrim/adhesive/2 mil Kapton[®]).
(a) After e^-/p^+ radiation exposure. (b) After e^-/p^+ radiation exposure, x-ray exposure
and thermal cycling.



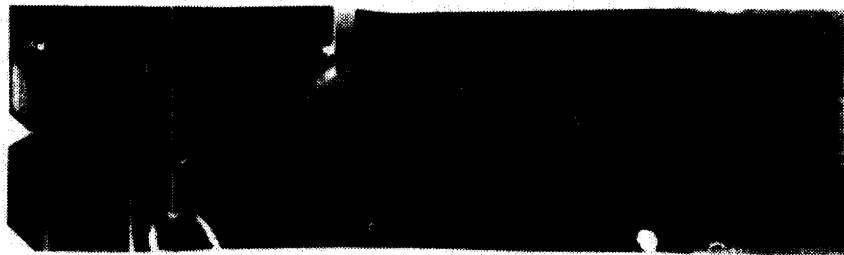
(a)



(b)



Figure 7. Candidate sample B1.6 (5 mil FEP/Al/adhesive Nomex[®] scrim). (a) After e^-/p^+ radiation exposure. (b) After e^-/p^+ radiation exposure, x-ray exposure and thermal cycling.



(a)



(b)

Figure 8. Candidate witness sample B1.8 (5 mil FEP/Al). (a) After e^-/p^+ radiation exposure.
(b) After e^-/p^+ radiation exposure, x-ray exposure and thermal cycling.

Generally, there were very small decreases in mass with thermal cycling ($<0.047\%$). These results are listed in Table 3. The optical properties for the samples after thermal cycling are listed in Table 4. Thermal cycling caused significant absorbance increases in sample B1.1 (+0.063) and sample B1.5 (+0.041). These samples have areas which appear yellowed with thermal cycling. Both of these samples have Ag and the Nomex[®] (vs. fiberglass) scrim. Sample B1.8 had an absorbance increase of +0.008 with thermal cycling. Samples B1.3 and B1.6 had slight decreases in absorbance (-0.005 and -0.007 , respectively). Both these samples have Al and the Nomex[®] scrim. Samples B1.2 and B1.4 (Ag and Al, respectively) with fiberglass scrims had essentially no absorbance change.

Optical microscopy provided evidence that the yellowed area of samples B1.1 and B1.5 are associated with the adhesive. Figure 9a shows that the yellowed area has a distinct pattern. Figure 9b shows the same area, photographed with back-lighting. Figure 9c shows the backside of the sample. It can be seen in Figure 9c that the polyester adhesive in the rectangular regions in between the fibers has occasionally cured with adhesive-voids.

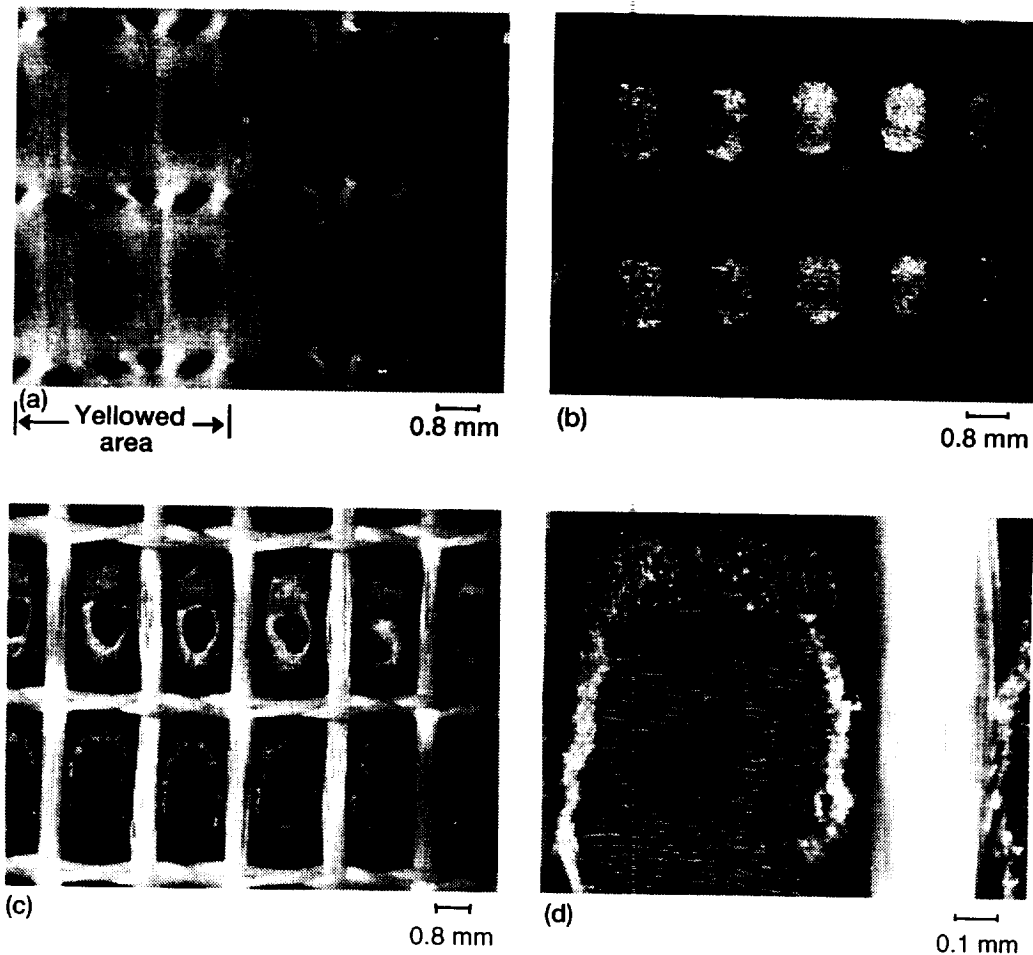


Figure 9. Candidate sample B1.5 after thermal cycling showing discoloration pattern. (a) Yellowed discolored area. (b) Same area as 9(a) with back lighting. (c) Backside of sample showing adhesive void areas. (d) Close-up of an adhesive void (cracks are in metal layers).

Table 5. Initial Cut Lengths and Tear Lengths after Thermal Cycling

Sample Id.	Initial Cut Length (mm)	Tear Length (mm)	Candidate Material
B1.1	10.2 (0.40")	1.8 (0.07")	10 mil FEP/Ag/Inconel/adhesive/Nomex® scrim
B1.2	9.1 (0.36")	28.7 (1.13")*	5 mil FEP/Ag/Inconel/adhesive/fiberglass scrim/adhesive/2 mil Kapton®
B1.3	9.4 (0.37")	5.3 (0.21")	10 mil FEP/Al/adhesive/Nomex® scrim
B1.4	9.1 (0.36")	28.7 (1.13")*	5 mil FEP/Al/adhesive/fiberglass scrim/adhesive/2 mil Kapton®
B1.5	8.9 (0.35")	2.8 (0.11")	5 mil FEP/Ag/Inconel/adhesive/Nomex® scrim
B1.6	9.1 (0.36")	5.1 (0.20")	5 mil FEP/Al/adhesive/Nomex® scrim
B1.7	9.9 (0.39")	5.3 (0.21")	OCLI multi-layer oxide UV blocker/2 mil white Tedlar®
B1.8	9.4 (0.37")	24.6 (0.97")	5 mil FEP/Al (current HST material)

* Sample tore in half

A close-up of such a void in the adhesive is shown in Figure 9d. The fine lines are cracks in the metal layers due to thermal cycling. Looking at the yellow pattern in 9a and the pattern of transmitted light in 9b associated with adhesive-voids, it becomes apparent that the discoloration is associated with the adhesive. It is not clear why the discoloration was not across the entire sample. As previously mentioned, both the discolored samples have Ag and Nomex® scrim and were supplied by Sheldahl with their non-UV darkening adhesive (FEP/Ag/Inconel/adhesive/Nomex® scrim). The Ag sample supplied by Sheldahl with a fiberglass scrim and Kapton® substrate (B1.2) did not discolor even though it had the same adhesive (FEP/Ag/Inconel/adhesive/fiberglass scrim/adhesive/Kapton®). The Al samples provided by Dunmore with Al and Nomex® scrim (B1.3 and B1.6) did not yellow with thermal cycling (FEP/Al/adhesive/Nomex® scrim). It therefore appears that there is a discoloring problem with the Sheldahl adhesive and Nomex® scrim upon heating.

Optical microscopy was used to document the extent of damage which occurred at the purposely cut area with thermal cycling under load. The initial cut lengths and tear lengths propagated during thermal cycling are listed in Table 5. The fiberglass/Kapton® samples (B1.2 and B1.4) tore in half. The tear lengths for the Nomex® scrim samples ranged from 1.8 mm (0.071") to 5.3 mm (0.209"). The Ag/Inconel Nomex® samples (B1.1 and B1.5) had shorter tears than the Al Nomex® samples (B1.3 and B1.6) of the same FEP thickness. The Nomex® scrim on the Al samples was not as orthogonal as the Ag/Inconel samples (as can be seen by comparing B1.1 and B1.6 in Figure 5) which may have contributed to the small additional tearing. Sample B1.7 had a very fine hairline crack which was 5.3 mm long, and the witness Al-FEP sample (B1.8) had a tear length of 42.6 mm. Figure 10 shows tearing of the cut tip in sample B1.6. The backside of the sample shown in Figure 10b shows the tear extending to the second fiber bundle, with no damage to the uncut fiber bundles near the propagated tear.

Several tests were run to understand the exact mechanism responsible for the tear propagation of the thermal cycled samples. Because the tensile properties of FEP decrease with temperature, loaded samples were exposed to the high end temperature (50°C) of the



Figure 10. Candidate sample B1.6 after thermal cycling. (a) Close up of purposely cut region showing fine tear which propagated during thermal cycling. (b) Backside of sample showing the tear extending to the second fiber bundle beyond the cut.

thermal cycling range to see if tearing which occurred with thermal cycling was due to decreases in tensile strength with temperature. Four 5 mil Al-FEP samples with identical sample dimensions and purposely cut marks as the candidate samples were loaded under tension (2.5, 5.0, 7.5 and 10 lb. loads) and exposed to 50°C for 90.5 hours (approximately the time the candidate samples were thermal cycled). There was no observed propagation of the cut with exposure under any of the applied loads. Al-FEP samples which were exposed to 50°C under load were then brought to liquid N₂ temperature while under tension to see if the additional load due to contraction caused tear propagation. There was no observed propagation of the cut with exposure under any of the applied loads.

Four Al-FEP (5 mil) samples, which had not been previously exposed to electron/proton or x-ray exposures, were thermal cycled under varying loads (2.5, 5.0, 7.5 and 10.0 lbs.). These samples also had identical geometry and cut marks as the candidate samples. After 250 cycles, the facility was stopped and opened up. No signs of tear propagation were observed at 250 cycles for any of the samples. After 1,000 cycles, small or no tears were observed for the 2.5, 5.0 and 7.5 lb. loaded samples, while the 10 lb. loaded sample was torn almost as far as candidate sample B1.8. Because of the thermal cycling test set-up, additional loads are imposed on the samples as they move from the hot to the cold chamber, and vice-versa. Therefore, another set of 5 mil Al-FEP samples were prepared and mechanically cycled under load (2.5, 5.0, 7.5 and 10 lb. loads) in the thermal cycling facility (at room temperature). After 250 mechanical cycles, the 2.5 and 5.0 lb. samples had no tear initiation, while the 7.5 and 10 lb. samples had very small tear initiations. After 1000 cycles, there was no tear propagation in either the 7.5 or the 10.0 lb. samples. These results indicate that the tearing which occurred in sample B1.8 was due to thermal cycling (versus thermal exposure) under the high load (10 lb.). The prior radiation exposure does not appear to have contributed to tear propagation in samples B1.8 because radiation exposure was not necessary to cause tear propagation. Also, mechanical cycling under load was not sufficient without thermal cycling to cause tear propagation.

The cracks, which propagated on HST, were found to have very smooth surfaces. Figure 11 shows examples of these smooth, featureless crack morphologies. Figure 11a is an electron micrograph of an in-space-propagated crack from the LS MLI sample retrieved during SM2. Similar smooth crack surfaces were found on cracks located near cable holes in the magnetometer MLI retrieved during SM1, as shown in Figure 11b and 11c. These smooth cracks are believed to have occurred due to slow crack growth under low stress in the presence of a degrading environment.⁹ Similar findings were observed for the SADA MLI retrieved during SM1, and have been described as being similar to stress-cracking of glassy materials.⁴ When the embrittled SM2 MLI sample is purposely cracked by bending the FEP space exposed surface in tension, the crack surface is not smooth and glassy in appearance, but more fibrous as seen in Figure 12. This also supports the idea that the smooth cracks occurred slowly in the space environment. The Al-FEP sample which was thermal cycled under a 10 lb. load, with no prior radiation exposure, had a very different surface morphology. Its crack surface has wave-like striations, as seen in Figure 13. This surface is much more ductile than the glassy-looking surfaces which cracked in space. The crack morphology of this thermal cycled sample is identical to that of sample B1.8, which did receive prior radiation exposure.⁹ The additional radiation exposures (electron/proton and x-ray) did not affect the crack propagation morphology in this sample. The smooth

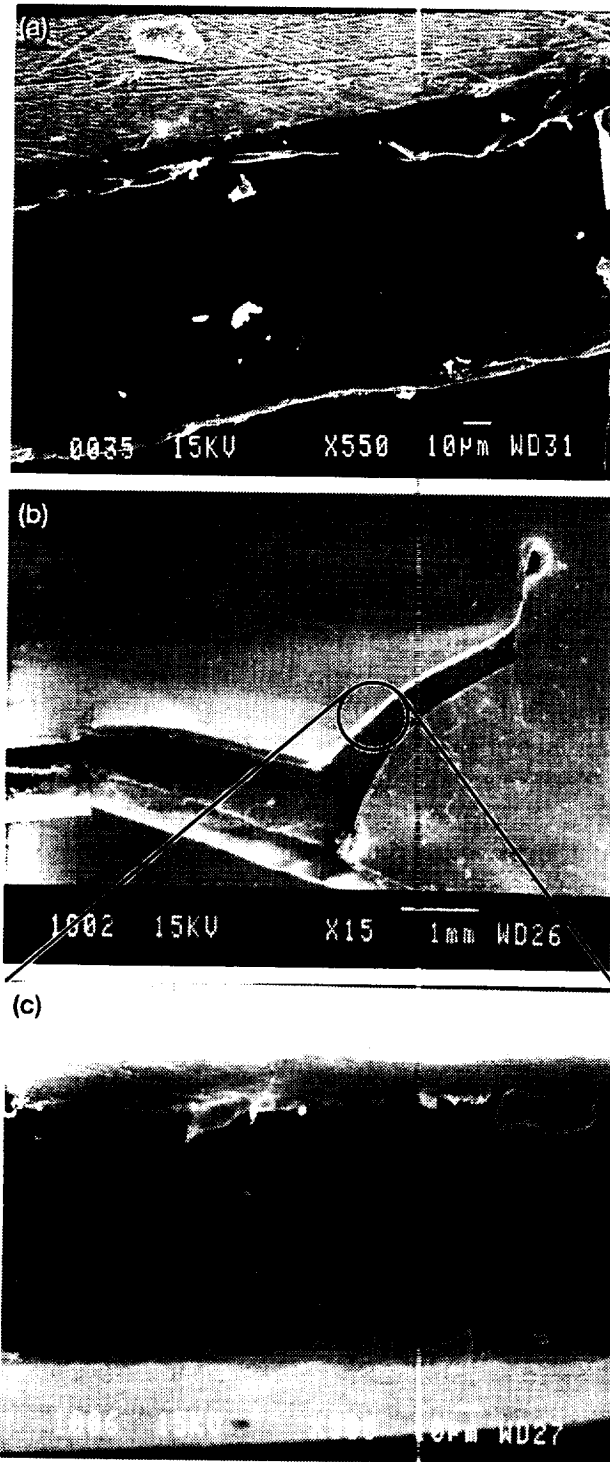


Figure 11. Smooth crack morphologies of in-situ propagated cracks in HST FEP. (a) SM2 LS FEP. (b) SM1 MSS FEP near a cable hole. (c) Close-up of crack surface in 11(b).



Figure 12. SM2 FEP after bending the space exposed FEP surface in tension. The tension induced fast-propagated crack has a fibrous, textured surface morphology.

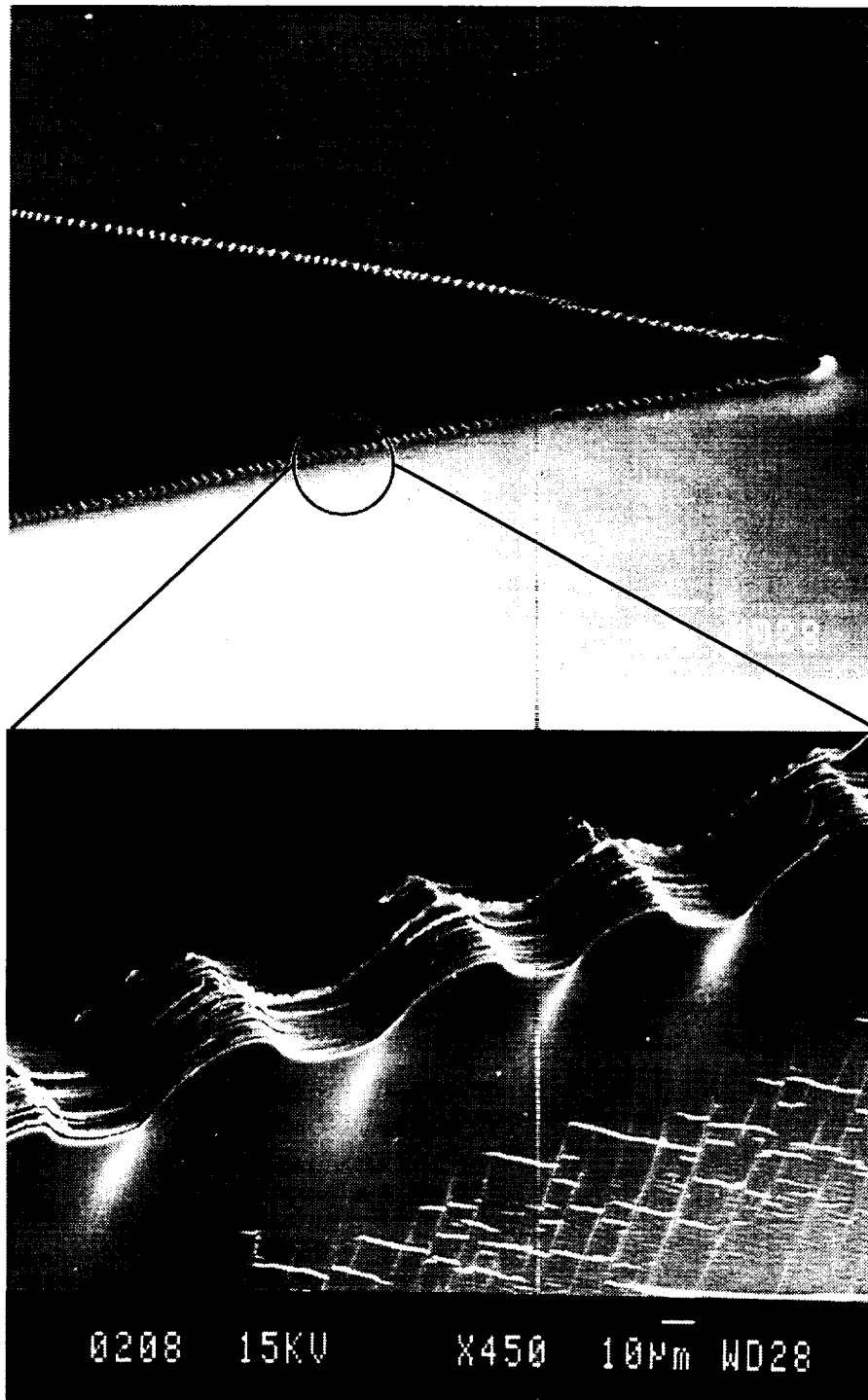


Figure 13. Surface morphology of FEP tear propagated during thermal cycling under load. This surface has ductile appearing, wave-like striations.

crack morphology which occurred on HST was not observed for the B1 samples which were thermal cycled under tensile loading, although one sample with scrim backing (B1.5) had areas which were relatively smooth.⁹ However, relatively smooth crack morphology was observed with most of the GSFC tested samples containing scrim which were thermal cycled under constraint loading (taped at edges) and experienced crack propagation.⁹

Based on the environmental testing conducted at LeRC, candidate samples B1.3 and B1.6 (Al and Nomex[®] scrim, prepared by Dunmore) performed the best. Samples B1.2 and B1.4 (with fiberglass scrim and Kapton[®] substrates) tore in half during thermal cycling under load. Samples B1.1 and B1.5 (with Ag and Nomex[®] scrim, prepared by Sheldahl) had significant increases in solar absorptance with thermal cycling which were attributed to an interaction of the adhesive and the Nomex[®] scrim with heating. Sample B1.7 (an OCLI coated sample) has a high initial absorptance, and was tested because the HST project wanted to check its performance. Because the solar absorptance of B1.6 was lower than B1.3, the HST MLI candidate sample B1.6 was considered the single best performer (using solar absorptance as the determining factor between B1.3 and B1.6).

4. Additional Testing

After sample characterization at LeRC (mass, optical properties and optical microscopy) the candidate samples were returned to GSFC for further testing and/or analyses. The M2 sample set was thermal cycled under constraint loading at GSFC and then exposed to NUV radiation. Other candidate sample sets were exposed to environments including electron/proton radiation, atomic oxygen and thermal cycling (sample set M1), thermal cycling of large samples (samples set L1) and NUV radiation (sample set G1).⁹

5. Final Ranking

When testing of all candidate samples was completed, the HST MLI FRB met to review the results and revote on the candidate materials using the original multiplicative evaluation formula based on the original nine performance criteria. During the intervening months, the HST project decided to use the sample 9 material (SiO₂/Al₂O₃/Ag/Al₂O₃/stainless steel) as the outer layer on all Bay repairs. The HST MLI FRB was then evaluating the test results for use on the HST LS and FS areas only. Based on this criteria, two of the original 10 materials (materials 9 and 10) were not considered in the final evaluation. Material 9 is not practical for the repairs because of handling (stainless steel sheets), and material 10 had problems with particulate contamination and UV darkening.⁹

The final ranking of the candidate materials is listed in Table 6. Material 6 (5 mil FEP/Ag/adhesive/Nomex[®] scrim) was ranked first, and recommended by the FRB as the replacement thermal control material to be installed on HST during SM3. Because of concerns of UV darkening of the adhesive, some of the material 6 samples underwent additional NUV exposure testing at GSFC. Five samples, were exposed for 2000-3000 ESH. No change in solar absorptance was measured.

Table 6. Final Ranking of HST Thermal Control Candidate Replacement Materials

Rank*	Material	Candidate Material
1	6	5 mil FEP/Al/adhesive/Nomex [®] scrim
2	3	10 mil FEP/Al/adhesive/Nomex [®] scrim
3	8	5 mil FEP/Al (current HST material)
4	1	10 mil FEP/Ag/Inconel/adhesive/Nomex [®] scrim
5	5	5 mil FEP/Ag/Inconel/adhesive/Nomex [®] scrim
6	2	5 mil FEP/Ag/Inconel/adhesive/fiberglass scrim/adhesive/2 mil Kapton [®]
7	7	OCLI multi-layer oxide UV blocker/2 mil white Tedlar [®]
8	4	5 mil FEP/Al/adhesive/fiberglass scrim/adhesive/2 mil Kapton [®]

* 1 is the most favorable candidate

6. Summary and Conclusions

The outer layer of MLI (5 mil Al-FEP) on HST is degrading in the space environment. A HST MLI FRB was chartered to determine the cause of degradation of the FEP, and to recommend a replacement material. The replacement thermal control material was required to meet HST's stringent thermal requirements (end-of-life $\alpha_p/\epsilon \leq 0.28$) and to maintain its structural integrity for at least 10 years on-orbit. Candidate replacement materials were chosen through a multiplicative evaluation formula based on nine performance criteria for durability testing. Various sets of candidate replacement materials were exposed to combinations of electron/proton radiation, atomic oxygen, simulated solar flare x-rays, thermal cycling and near ultraviolet radiation at various facilities in order to evaluate their HST on-orbit durability. Two sets of samples (B1 and M2) previously exposed to charged particle radiation were exposed to x-rays at LeRC. One sample set (B1) was also thermal cycled under load at LeRC.

There were very small decreases in mass, and no changes in optical properties, and no change in the original cut, of the candidate materials with simulated solar flare x-ray exposure to 10 year HST fluences (SM3-2010). Small decreases were observed in the tensile properties of companion 5 mil Al-FEP witness samples with x-ray exposure. Generally, there were very small changes in mass with thermal cycling. Thermal cycling caused significant absorptance increases in samples B1.1 (+0.063) and B1.5 (+0.041) which were attributed to an interaction of the Nomex[®] scrim with the Sheldahl adhesive upon heating. Samples B1.3 and B1.6, with Al and Nomex[®] scrims had absorptance decreases with thermal cycling (-0.005 and -0.007, respectively). Samples B1.2 and B1.4, with fiberglass scrims and Kapton[®] substrates, tore in half during thermal cycling under load. Sample B1.8 (current HST MLI material) tore about 90 percent of the width during thermal cycling. Tear propagation of the B1 samples was attributed to thermal cycling under a high load. The prior radiation exposures did not appear to have an additional effect on tearing, and no tearing occurred due to mechanical load cycling. The crack morphology of sample B1.8 (5 mil Al-FEP) exposed to radiation then thermal cycled under a high load (10 lb.)

was ductile in appearance, unlike the glassy crack morphology of the brittle FEP from HST. The crack morphology of radiation exposed scrim containing samples thermal cycled under a low load (constraint loading) at GSFC most closely resembled in-space-propagated cracks from retrieved HST FEP. Based on x-ray and thermal cycling testing at LeRC, samples B1.3 and B1.6 performed the best, with B1.6 having a lower solar absorptance.

Upon completion of testing all candidate samples, the HST MLI FRB met to review the durability results and revote on the candidate materials using the original multiplicative evaluation formula. Material 6 (5 mil FEP/Al/adhesive/Nomex[®] scrim) was ranked first, and recommended by the FRB as the replacement thermal control material to be installed on HST during SM3.

7. Acknowledgments

The authors would like to thank the following for their contributions to this program: Tom Stueber of NYMA Inc., and Alice Jalics of Cleveland State University for tensile tests, Kevin Malinowski of Cleveland State University for assistance with thermal cycling, and Dr. Steven Pepper of NASA Lewis Research Center for x-ray spectrum characterization. We would also like to thank Patti Hansen (the HST MLI FRB Chair) and Jackie Townsend of NASA Goddard Space Flight Center, and Joyce Dever of NASA Lewis Research Center for helpful discussions.

8. References

1. Hansen, Patricia A., Townsend, Jacqueline A., Yoshikawa Yukio, Castro, David J., Triolo, Jack J. and Peters, Wanda C., "Degradation of Hubble Space Telescope Metallized Teflon[®] FEP Thermal Control Materials." Science of Advanced Materials and Process Engineering Series, 43, 1998.
2. Zuby, Thomas M., de Groh, Kim K. & Smith, Daniela C., "Degradation of FEP Thermal Control Materials Returned from the Hubble Space Telescope." Proceedings of the Hubble Space Telescope Solar-Array Workshop, Noordwijk, The Netherlands, May 30–31, 1995, ESA WPP-77, pp. 385–402; also NASA TM–104627, December 1995.
3. de Groh, Kim K. & Smith, Daniela C., "Investigation of Teflon[®] FEP Embrittlement on Spacecraft in Low Earth Orbit." Proceedings of the 7th International Symposium on Materials in Space Environment, Toulouse, France, June 16–20, 1997, ESA SP-399; also NASA TM–113153, 1997.
4. Van Eesbeek, Marc, Levadou, Francois & Milintchouk, Andrei, "Investigation on FEP from PDM and Harness from HST-SA1." Proceedings of the Hubble Space Telescope Solar-Array Workshop, Noordwijk, The Netherlands, May 30–31, 1995, ESA WPP-77, pp. 403–416.
5. Townsend, Jacqueline A., Hansen, Patricia A., Dever, Joyce A. and Triolo, Jack J., "Analysis of Retrieved Hubble Space Telescope Thermal Control Materials." Science of Advanced Materials and Process Engineering Series, 43, 1998.
6. Dever, Joyce A., de Groh, Kim K., Townsend, Jacqueline A. and Wang, L. Len, "Mechanical Properties Degradation of Teflon[®] FEP Returned From the Hubble Space Telescope." Presented at the 36th Aerospace Sciences Meeting & Exhibit, Reno, NV, January 12-15, 1998, AIAA-98-0895, NASA TM–1998-206618.
7. Dever, Joyce A., Townsend, Jacqueline A., Gaier, James R. and Jalics, Alice I., "Synchrotron VUV and Soft X-Ray Radiation Effects on Aluminized Teflon[®] FEP." Science of Advanced Materials and Process Engineering Series, 43, 1998.

8. Banks, Bruce A., de Groh, Kim K., Stueber, Thomas J. and Sochkar, Edward A., "Ground Laboratory Soft X-Ray Durability Evaluation of Aluminized Teflon[®] FEP Thermal Control Insulation." Science of Advanced Materials and Process Engineering Series, 43, 1998.
9. Townsend, Jacqueline A., Hansen, Patricia A., McClendon, Mark W., Dever, Joyce A. and Triolo, Jack J., "Evaluation and Selection of Replacement Thermal Control Materials for the Hubble Space Telescope." Science of Advanced Materials and Process Engineering Series, 43, 1998.
10. Henninger, John H., "Solar Absorptance and Thermal Emittance of Some Common Spacecraft Thermal-Control Coatings." NASA RP-1121, 1984.
11. Edwards, David L. and Vaughn, Jason A., "Charged Particle and Atomic Oxygen Exposure of Candidate Replacement Materials for the Hubble Space Telescope," NASA MSFC Test Report, October 15, 1997.
12. The Boeing Company, "Radiation Exposure Testing—HST Thermal Blanket," Boeing Test Report, November, 1997.
13. Milintchouk, Andrei, Van Eesbeek, Marc, Levadou, Francois and Harper, Tim, "Influence of X-Ray Solar Flare Radiation on Degradation of Teflon[®] in Space." J. of Spacecraft and Rockets, Vol. 34, No. 4, July-August 1997, pp. 542-548.

REPORT DOCUMENTATION PAGE

Form Approved
OMB No. 0704-0188

Public reporting burden for this collection of information is estimated to average 1 hour per response, including the time for reviewing instructions, searching existing data sources, gathering and maintaining the data needed, and completing and reviewing the collection of information. Send comments regarding this burden estimate or any other aspect of this collection of information, including suggestions for reducing this burden, to Washington Headquarters Services, Directorate for Information Operations and Reports, 1215 Jefferson Davis Highway, Suite 1204, Arlington, VA 22202-4302, and to the Office of Management and Budget, Paperwork Reduction Project (0704-0188), Washington, DC 20503.

1. AGENCY USE ONLY (Leave blank)	2. REPORT DATE December 1998	3. REPORT TYPE AND DATES COVERED Technical Memorandum	
4. TITLE AND SUBTITLE Simulated Solar Flare X-Ray and Thermal Cycling Durability Evaluation of Hubble Space Telescope Thermal Control Candidate Replacement Materials		5. FUNDING NUMBERS WU-632-1A-1E-00	
6. AUTHOR(S) Kim K. de Groh, Bruce A. Banks, Edward A. Sechkar, and David A. Scheiman			
7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES) National Aeronautics and Space Administration Lewis Research Center Cleveland, Ohio 44135-3191		8. PERFORMING ORGANIZATION REPORT NUMBER E-11184	
9. SPONSORING/MONITORING AGENCY NAME(S) AND ADDRESS(ES) National Aeronautics and Space Administration Washington, DC 20546-0001		10. SPONSORING/MONITORING AGENCY REPORT NUMBER NASA TM-1998-207426	
11. SUPPLEMENTARY NOTES Prepared for the Fourth International Space Conference on Protection of Materials and Structures from the LEO Space Environment sponsored by the University of Toronto and the Institute for Aerospace Studies, Toronto, Canada. April 23-24, 1998. Kim K. de Groh and Bruce A. Banks, NASA Lewis Research Center; Edward A. Sechkar and David A. Scheiman, NYMA, Inc., 2001 Aerospace Parkway, Brook Park, Ohio 44142. Responsible person, Kim K. de Groh, organization code 5480, (216) 433-2297.			
12a. DISTRIBUTION/AVAILABILITY STATEMENT Unclassified - Unlimited Subject Categories: 18, 92, and 27 This publication is available from the NASA Center for AeroSpace Information, (301) 621-0390.		12b. DISTRIBUTION CODE Distribution: Nonstandard	
13. ABSTRACT (Maximum 200 words) During the Hubble Space Telescope (HST) second servicing mission (SM2), astronauts noticed that the multilayer insulation (MLI) covering the telescope was damaged. Large pieces of the outer layer of MLI (aluminized Teflon [®] fluorinated ethylene propylene (Al-FEP)) were torn in several locations around the telescope. A piece of curled up Al-FEP was retrieved by the astronauts and was found to be severely embrittled, as witnessed by ground testing. Goddard Space Flight Center (GSFC) organized a HST MLI Failure Review Board (FRB) to determine the damage mechanism of FEP in the HST environment, and to recommend replacement insulation material to be installed on HST during the third servicing mission (SM3) in 1999. Candidate thermal control replacement materials were chosen by the FRB and tested for environmental durability under various exposures and durations. This paper describes durability testing of candidate materials which were exposed to charged particle radiation, simulated solar flare x-ray radiation and thermal cycling under load. Samples were evaluated for changes in solar absorptance and tear resistance. Descriptions of environmental exposures and durability evaluations of these materials are presented.			
14. SUBJECT TERMS Hubble space telescope; Thermal control coatings; Solar x-rays; Absorptance; Teflon; Thermal cycling; Space environment simulation; Soft x-rays		15. NUMBER OF PAGES 32	
		16. PRICE CODE A03	
17. SECURITY CLASSIFICATION OF REPORT Unclassified	18. SECURITY CLASSIFICATION OF THIS PAGE Unclassified	19. SECURITY CLASSIFICATION OF ABSTRACT Unclassified	20. LIMITATION OF ABSTRACT

