



Ion Species Fractions in the Far-Field Plume of a High-Specific Impulse Hall Thruster

Richard R. Hofer
QSS Group, Inc., Cleveland, Ohio

Alec D. Gallimore
University of Michigan, Ann Arbor, Michigan

The NASA STI Program Office . . . in Profile

Since its founding, NASA has been dedicated to the advancement of aeronautics and space science. The NASA Scientific and Technical Information (STI) Program Office plays a key part in helping NASA maintain this important role.

The NASA STI Program Office is operated by Langley Research Center, the Lead Center for NASA's scientific and technical information. The NASA STI Program Office provides access to the NASA STI Database, the largest collection of aeronautical and space science STI in the world. The Program Office is also NASA's institutional mechanism for disseminating the results of its research and development activities. These results are published by NASA in the NASA STI Report Series, which includes the following report types:

- **TECHNICAL PUBLICATION.** Reports of completed research or a major significant phase of research that present the results of NASA programs and include extensive data or theoretical analysis. Includes compilations of significant scientific and technical data and information deemed to be of continuing reference value. NASA's counterpart of peer-reviewed formal professional papers but has less stringent limitations on manuscript length and extent of graphic presentations.
- **TECHNICAL MEMORANDUM.** Scientific and technical findings that are preliminary or of specialized interest, e.g., quick release reports, working papers, and bibliographies that contain minimal annotation. Does not contain extensive analysis.
- **CONTRACTOR REPORT.** Scientific and technical findings by NASA-sponsored contractors and grantees.

- **CONFERENCE PUBLICATION.** Collected papers from scientific and technical conferences, symposia, seminars, or other meetings sponsored or cosponsored by NASA.
- **SPECIAL PUBLICATION.** Scientific, technical, or historical information from NASA programs, projects, and missions, often concerned with subjects having substantial public interest.
- **TECHNICAL TRANSLATION.** English-language translations of foreign scientific and technical material pertinent to NASA's mission.

Specialized services that complement the STI Program Office's diverse offerings include creating custom thesauri, building customized databases, organizing and publishing research results . . . even providing videos.

For more information about the NASA STI Program Office, see the following:

- Access the NASA STI Program Home Page at **<http://www.sti.nasa.gov>**
- E-mail your question via the Internet to **help@sti.nasa.gov**
- Fax your question to the NASA Access Help Desk at 301-621-0134
- Telephone the NASA Access Help Desk at 301-621-0390
- Write to:
NASA Access Help Desk
NASA Center for Aerospace Information
7121 Standard Drive
Hanover, MD 21076



Ion Species Fractions in the Far-Field Plume of a High-Specific Impulse Hall Thruster

Richard R. Hofer
QSS Group, Inc., Cleveland, Ohio

Alec D. Gallimore
University of Michigan, Ann Arbor, Michigan

Prepared for the
39th Joint Propulsion Conference and Exhibit
cosponsored by AIAA, ASME, SAE, and ASEE
Huntsville, Alabama, July 20-23, 2003

Prepared under Contract NAS3-00145

National Aeronautics and
Space Administration

Glenn Research Center

Acknowledgments

Program support for this research through Code R Energetics funding is gratefully acknowledged.

Available from

NASA Center for Aerospace Information
7121 Standard Drive
Hanover, MD 21076

National Technical Information Service
5285 Port Royal Road
Springfield, VA 22100

Available electronically at <http://gltrs.grc.nasa.gov>

ION SPECIES FRACTIONS IN THE FAR-FIELD PLUME OF A HIGH-SPECIFIC IMPULSE HALL THRUSTER

Richard R. Hofer
QSS Group, Inc.
Cleveland, Ohio 44135
richard.hofer@grc.nasa.gov

Alec D. Gallimore
University of Michigan
Plasmadynamics and Electric Propulsion Laboratory
Ann Arbor, Michigan 48109

ABSTRACT

An ExB probe was used to measure the ion species fractions of Xe^+ , Xe^{2+} , and Xe^{3+} in the far-field plume of the NASA-173Mv2 laboratory-model Hall thruster. The thruster was operated at a constant xenon flow rate of 10 mg/s and discharge voltages of 300 to 900 V. The ExB probe was placed two meters downstream of the thruster exit plane on the thruster centerline. At a discharge voltage of 300 V, the species fractions of Xe^{2+} and Xe^{3+} were lower, but still consistent with, previous Hall thruster studies using other mass analyzers. Over discharge voltages of 300 to 900 V, the Xe^{2+} species fractions increased from 0.04 to 0.12 and the Xe^{3+} species fraction increased from 0.01 to 0.02.

I. INTRODUCTION

As used for spacecraft applications in Earth orbit such as station-keeping, orbit-raising, and orbit-transfers, the xenon Hall thruster is most often regarded as a 1600 second specific impulse device operating at discharge voltages of 300 V. Mission studies have shown though that increases in the specific impulse of Hall thrusters to 2000 to 3000 seconds can enhance or enable a number of Earth-orbital and interplanetary missions.¹⁻⁴

The NASA-173M series of laboratory-model Hall thrusters were conceived in order to understand the design challenges and physical mechanisms determining performance at high-specific impulse (2000 to 3000 seconds).⁵⁻⁹ The basic design philosophy was that changes to the magnetic field topography would be required for efficient

operation above 1600 seconds specific impulse. Advanced magnetic circuits allowing for *in situ* variation of the magnetic field topography were therefore incorporated into each thruster. The NASA-173Mv1, built jointly by the University of Michigan and the NASA Glenn Research Center (GRC), first established the validity of this approach.^{5,6} In a performance study at low current density, changes to the magnetic field topography at high-specific impulse were shown to be critical to achieving efficient operation.⁶ The follow-on version, the GRC-developed NASA-173Mv2, incorporated design improvements suggested by its predecessor. A performance mapping of the NASA-173Mv2 at several current densities established that efficient operation at high-specific impulse was possible if a minimum current density was maintained and the magnetic field topography was properly shaped.⁷

Design and experiment have established that Hall thrusters can operate efficiently at high-specific impulse. A need still exists though to improve understanding of the relationship between the thruster design and the microscopic plasma properties. This is necessary to further increase performance, evaluate and improve lifetime, and enable integration with spacecraft. To those ends, a series of plasma diagnostics have been used with the NASA-173Mv2 to better understand the plasma characteristics.⁷⁻⁹

In the present work, an ExB probe was used to measure the variation of the ion species fractions with discharge voltage in the NASA-173Mv2 Hall thruster. An ExB probe is a plasma diagnostic that

selectively filters charged particles according to their velocities.^{10–21} Because multiply-charged ions in Hall thrusters will have velocities proportional to the square root of their charge-state ($q^{1/2}$),²² an ExB probe can discriminate between distinct ion species. Examination of the relative height of the ion current peaks from the probe can then be used to compute the species fractions.

Measurements of the species fractions are important because multiply-charged ions affect thruster performance and the lifetime of the thruster and sensitive spacecraft surfaces (e.g., solar arrays and optics). Increased erosion by multiply-charged ions (due to their increased energy) can also decrease the lifetime of the thruster and spacecraft surfaces. At high discharge voltages, the importance of multiply-charged ions on Hall thruster performance and lifetime is relatively unknown because measurements have thus far been limited to 300 to 500 V in xenon Hall thrusters.^{19,21,23–26}

II. EXPERIMENTAL APPARATUS

The thruster hardware and configuration were the same as the experiments reported in reference 7. This included the thruster, power electronics, vacuum facility, and the thruster mounting scheme in the vacuum facility. In the experiments reported here, an ExB probe was added two meters downstream of the thruster exit plane on the thruster centerline.

A. NASA-173Mv2 Hall Effect Thruster

The NASA-173Mv2 is a 5 kW-class laboratory-model Hall thruster that has a discharge chamber outer diameter of 173 mm (Figure 1; see reference 7 for a detailed description). A fixed structure of magnetic poles pieces, an inner coil (IC) and an outer coil (OC) are used to form the primary magnetic circuit. Fine control of the magnetic field is provided with an internal trim coil (ITC) and an external trim coil (ETC). In the data presented in this paper, only the inner and outer coils were used. The results of experiments including the effects of the internal and external trim coils on ion species fractions have not yet been analyzed.

Figure 2 shows the total specific impulse and total efficiency of the NASA-173Mv2 at 10 mg/s versus discharge voltage.⁷ The specific impulse and efficiency include the cathode flow and the efficiency also includes the power to the magnets. Specific impulse and efficiency range from 1600 to 3400 seconds and 51 to 61%, respectively, over voltages of 300 to 1000 V.

The plasma discharge was powered by a matching pair of commercially available power supplies wired in series that provided a 1200 V, 16 A output. The discharge filter consisted of a 100 μ F capacitor in parallel with the supply outputs. Other commercially available power supplies were used to power the magnet coils and the cathode heater and keeper. The laboratory-model hollow cathode was a GRC design capable of emission currents up to 20 A. As shown in Figure 1, the cathode was positioned above the thruster.

High-purity xenon was supplied through stainless steel feed lines with 20 and 200 sccm mass flow controllers. The controllers were calibrated before the experiments using a constant-volume method. Based on the calibrations, the uncertainty was estimated as $\pm 0.9\%$ for the anode flow rate and $\pm 1.1\%$ for the cathode flow rate.



Figure 1.—Photograph of the NASA-173Mv2 Hall thruster.

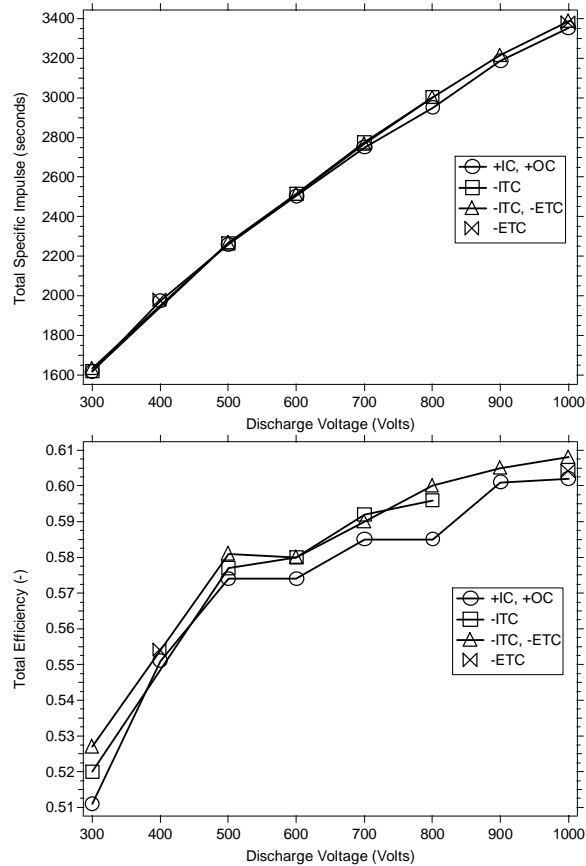


Figure 2.—Total specific impulse and efficiency versus discharge voltage of the NASA-173Mv2 at 10 mg/s. (from reference 7)

Thruster telemetry was acquired using a 22-bit datalogger. The DC accuracy of the unit, as reported by the manufacturer, is 0.004%. However, calibration of each channel using digital multimeters increased the uncertainty to $\pm 0.05\%$ for voltage and $\pm 0.2\%$ for current.

The thruster was operated for four hours after initial exposure to vacuum conditions to allow for outgassing of the chamber walls. Upon subsequent thruster shutdowns and restarts or a change in the discharge voltage, the thruster was operated for at least 30-60 minutes before data was acquired. This procedure allowed enough time for the discharge current to reach a steady-state value.

B. Vacuum Facility

The experiments were conducted in vacuum facility 12 (VF12) at GRC. VF12 is a cylindrical, stainless steel chamber 3.0 m in diameter by 9.6 m

in length. The facility is cryogenically pumped and backed by a turbomolecular pump for removal of gases not pumped by the cryosurfaces. The thruster was mounted on the thrust stand described in reference 7. The thrust stand pendulum was locked down and not used for these experiments. In this position, the thruster was near the chamber's vertical centerline and fired 8.9 m down the length of the tank toward the pumps, which are located along the back half of the chamber. A hot-cathode ionization gauge was mounted 0.4 m below the vertical chamber centerline, 5.2 m downstream of the thruster. Pressure measurements were corrected for xenon using the base pressure on air and a correction factor of 2.87 for xenon. For a total xenon flow rate of 11 mg/s, the pressure was 4.6×10^{-6} Torr, after correcting for xenon and the base pressure on air, which was 1.0×10^{-7} Torr. This test pressure corresponded to a xenon pumping speed of 340,000 l/s.

C. ExB Probe (Wien filter)

1. Theory of Operation

The classical dynamics of a beam of charged particles in electric and magnetic fields are governed by the Lorentz force equation

$$\vec{F} = qe(\vec{E} + \vec{u} \times \vec{B}) \quad (1)$$

(where the symbols have their usual meaning). The ExB probe, also known as a Wien filter, selects particles with a particular velocity by balancing the electric and magnetic fields such that there is no net force acting on those particles.¹⁰⁻²¹

$$0 = qe(\vec{E} + \vec{u} \times \vec{B}) \quad (2)$$

If the ExB probe is constructed with the electric and magnetic field perpendicular to each other and the particle velocity

$$\begin{aligned} \vec{E} &= (0, E, 0) \\ \vec{B} &= (0, 0, B) \\ \vec{u} &= (u, 0, 0) \end{aligned} \quad (3)$$

then Eq. (2) becomes

$$u = -\frac{E}{B} \quad (4)$$

Equation (4) shows that the particle velocity can be chosen by adjusting the electric and magnetic fields. In practice, a constant magnetic field is usually applied with permanent magnets. The electric field is usually established between two parallel plates separated by a gap, d , that are biased to a potential, V_{probe} , such that

$$E = -\frac{V_{probe}}{d} \quad (5)$$

Then the velocity pass condition in an ExB probe is

$$u = \frac{V_{probe}}{Bd} \quad (6)$$

Particles that satisfy Eq. (6) pass through the ExB field region and reach a collection electrode where they are recorded as current. The resulting current-voltage characteristic is therefore related to the velocity distribution function. It is important to note that an ExB probe is a velocity selector because the charge-state and mass of the ions do not appear in Eq. (4). Thus, an ExB probe will not detect signatures due to charge-exchange collisions but will detect elastic collisions through signal broadening.

In a Hall thruster, particles are accelerated electrostatically to a velocity given by²²

$$u_i = \sqrt{\frac{2q_i e V_{a,i}}{m_{xe}}} \quad (7)$$

where the subscript, i , indicates the charge of the ion ($i = 1, 2, 3, \dots$), $V_{a,i}$ is the effective accelerating potential, and m_{xe} is the mass of a xenon atom. Substituting Eq. (7) into (6) and solving for the plate voltage yields an expression in terms of the effective accelerating potential of each ion species and charge-state (i.e., the ion energy)

$$V_{probe,i} = \sqrt{\frac{2q_i e V_{a,i}}{m_{xe}}} (Bd) \quad (8)$$

The accelerating potential that each charge-state experiences in Hall thrusters differs only by a few tens of volts.^{19,21} Thus, the charge-states will roughly appear in an ExB current-voltage characteristic at multiples of $q^{1/2}$ above the singly-charged peak.

2. Design and Experimental Setup

The ExB probe was based on previous designs used to study ion thrusters.²⁰ Kim's ExB probe, which was the first to be used on a Hall thruster, was also based on these designs.^{19,21} Shown schematically in Figure 3, the probe used in these experiments consisted of three main sections: the entrance collimator, ExB test section, and exit collimator. The entrance collimator was 127 mm in length and had two circular orifices at either end that were 0.5 and 1.0 mm in diameter. In the 127 mm long test section, the magnetic field was applied with permanent magnets that provided a magnetic field strength at the test section center of 0.16 T. The electric field was established with a pair of aluminum plates machined from channel stock. The bias plates were separated by a distance of $d = 42.5$ mm with legs used to minimize electric field fringing that were $d/4$ in length.^{14,17} The exit collimator was 127 mm long and had an exit orifice diameter of 3.2 mm. A 23 mm diameter tungsten collection electrode was placed at the end of the exit collimator. The entrance and exit collimator orifices were aligned using a laser. The acceptance angle of the probe was 0.7° . Probe resolution was conservatively estimated as 7% of the ion energy.¹⁹

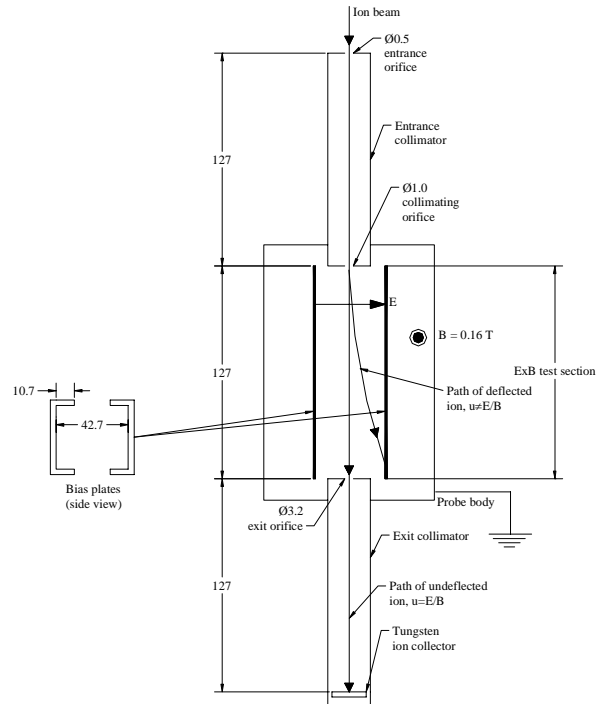


Figure 3.—Schematic of the ExB probe. Dimensions are given in millimeters.

The entrance orifice of the probe was placed two meters downstream of the thruster exit plane and on the thruster centerline, within an accuracy of ± 0.5 cm in both the axial and radial directions. The entrance and exit collimators were aligned perpendicular to the thruster exit plane with a laser to an accuracy of $\pm 0.5^\circ$.

Figure 4 shows the electrical schematic of the ExB probe. The plate bias was applied with a commercially available power supply and was swept over a range of voltages in two volt steps. For a given thruster operation condition, each voltage sweep was repeated 3 to 5 times and then averaged together to reduce measurement uncertainty. The supply outputs were biased with respect to ground using a pair of 1 M Ω resistors so that the potential at the mid-point of the plates was held near facility ground. Current from the collection electrode was measured with a picoammeter through 50 Ohm, shielded coaxial cable. Except for the bias plates and the collection electrode, the entire probe was grounded to the vacuum facility.

3. Data Analysis

Since the goal of the experiments was to obtain the ion species fractions as a function of the discharge voltage, data analysis consisted of interpreting the

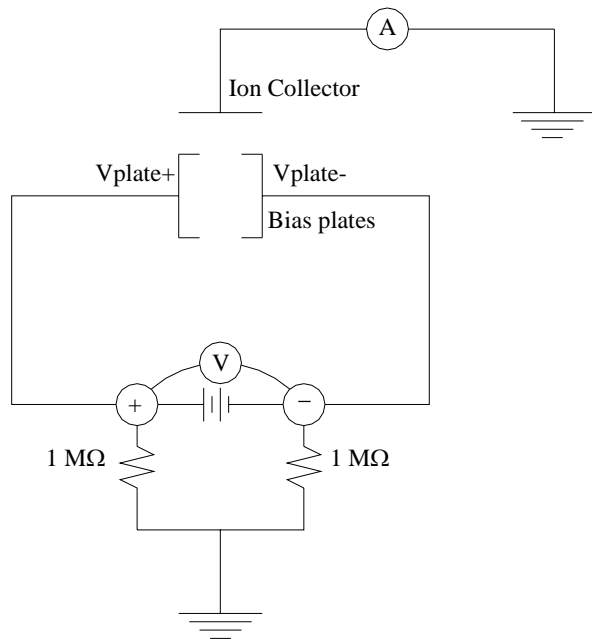


Figure 4.—Electrical schematic of the ExB probe.

peak heights measured by the ExB probe. The Xe^+ and Xe^{2+} peaks were easily identified for all operating conditions. Signal broadening due to elastic collisions and probe resolution limitations resulted in the Xe^{2+} and Xe^{3+} signals overlapping at discharge voltages of 500 to 800 V. In those cases, the Xe^{3+} ion current was taken where the first derivative of the ion current crossed, or nearly crossed, zero. The voltage where this occurred always closely corresponded to $3^{1/2} \cdot V_{\text{probe},1}$. Some discharge voltages showed evidence of Xe^{4+} , but the signal never represented a significant fraction of the total signal. For example, if the signal at 900 V (where Xe^{4+} was most evident, see Figure 7 in section III) were included in the species fractions calculations, then the Xe^{4+} fraction would have been 0.001. Because the signal was so small, Xe^{4+} was not included in the calculations.

Using the peak heights to calculate the species fractions was equivalent to approximating the ion energy distribution function as a monoenergetic beam. In a Hall thruster, the beam has a distribution with a finite width of energies, since ions are produced along the length of the discharge chamber at different accelerating potentials. Kim^{19,21} accounted for these effects by including a model of the ion energy distribution function and found reasonable agreement with other mass analyzers.^{23–26} While modeling the distribution is a more rigorous approach that should reduce uncertainty, it requires that probe-induced signal broadening is much less than collisional broadening. The ExB probe used in these experiments had an energy resolution of 7% (compared to 1% with Kim's probe), which was too high to neglect probe broadening. As a result, it was concluded that modeling the distribution would introduce more uncertainty to the calculations than using the peak heights. While using the peak heights introduces additional uncertainty, the computed species fractions were still comparable to other mass analyzers (see section III below). A conservative estimate of the uncertainty that included the effects of using the peaks heights is discussed in section II.C.4.

The ion current of each species from the probe traces was expressed as

$$I_i = eq_i n_i u_i A_c (1 + \gamma_i) \quad (9)$$

where A_c was the current collection area, γ_i was the secondary electron emission yield (given in electrons per ion), and n_i was the number density.

The secondary electron emission yield was taken from data in reference 27 for xenon bombardment on tungsten which showed that γ_i was on average 0.018, 0.216, and 0.756 for Xe^+ , Xe^{2+} , and Xe^{3+} , respectively, over the range of ion energies in these experiments.

From the ion currents, the current fractions were computed as

$$\Omega_i = \frac{I_i}{\sum I_i} \quad i = 1, 2, 3 \quad (10)$$

where

$$\sum \Omega_i = 1 \quad (11)$$

The ion species fractions were given by

$$\zeta_i = \frac{n_i}{\sum n_i} \quad i = 1, 2, 3 \quad (12)$$

where

$$\sum \zeta_i = 1 \quad (13)$$

Equations (7), (9), and (12) were then substituted into Eq. (10) to obtain an expression relating the current and species fractions

$$\Omega_i = \frac{q_i^{3/2} \zeta_i (1 + \gamma_i)}{\sum (q_i^{3/2} \zeta_i (1 + \gamma_i))} \quad (14)$$

where it was assumed that the collection area and effective acceleration potential of each species were the same. Solving the system of equations defined by Eqs. (13) and (14) yielded the ion species fractions.

4. Measurement Uncertainty

To estimate the measurement uncertainty, the sensitivity of each ion species fraction was analyzed by considering: the uncertainty in the probe voltage and current identified for each species, probe misalignment and resolution, the variation in accelerating potential with ion species, and whether the secondary electron emission yield

was included. Based on this analysis, the uncertainty in the species fractions was estimated to be ± 0.02 , ± 0.01 , and ± 0.005 for Xe^+ , Xe^{2+} , and Xe^{3+} , respectively. Other sources of error, such as the loss of particles over the two meter path length and the effects of using the peak heights to identify the species current fractions, were not included. Methods to estimate the added uncertainty of these effects are currently being considered. However, preliminary estimates did show that the additional uncertainty was on the order of the uncertainty from all other sources. As a result, the uncertainty was revised to the more conservative estimate of ± 0.04 , ± 0.02 , and ± 0.01 for Xe^+ , Xe^{2+} , and Xe^{3+} , respectively. At 300 V, this equated to a percentage uncertainty of $\pm 4\%$, $\pm 50\%$ and $\pm 100\%$ for Xe^+ , Xe^{2+} , and Xe^{3+} , respectively.

III. RESULTS AND DISCUSSION

ExB probe measurements were taken with the thruster operating at an anode flow rate of 10.0 mg/s and a cathode flow rate of 1.0 mg/s. The discharge voltage was varied from 300 to 900 V. The ExB probe was two meters downstream of the thruster exit plane on the thruster centerline. At each discharge voltage, the effects of the magnetic field were evaluated by using several combinations of the coils. In this paper, data with the inner and outer coils are presented. The results of experiments including the effects of the internal and external trim coils have not yet been analyzed. Thruster telemetry from the test series are in the appendix as Table 1.

Figures 5 to 7 are representative ExB probe traces at 300, 600, and 900 V, respectively. Each figure is the average of three to five voltage sweeps of the ExB probe. In general, the probe traces yielded distinct and repeatable spectra for the Xe^+ , Xe^{2+} , and Xe^{3+} peaks. The Xe^{2+} and Xe^{3+} peaks closely corresponded to the expected $q^{1/2}$ dependence.

The widths of the probe spectra for each species show a clear broadening with discharge voltage. This result correlated with measurements of the ion voltage distribution that showed the same trend.⁸ The implications of these results on thruster operation are discussed further in reference 8.

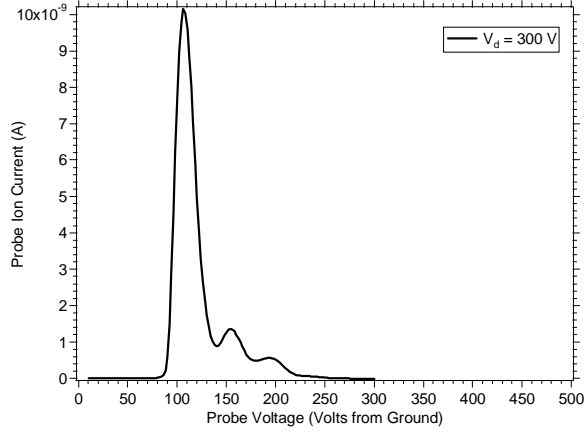


Figure 5.—ExB probe ion current versus probe voltage at 300 V.

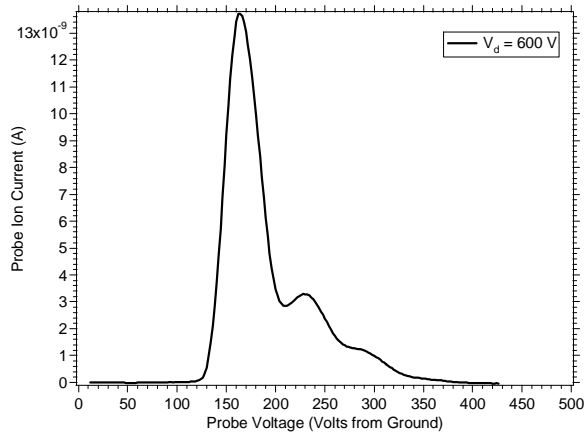


Figure 6.—ExB probe ion current versus probe voltage at 600 V.

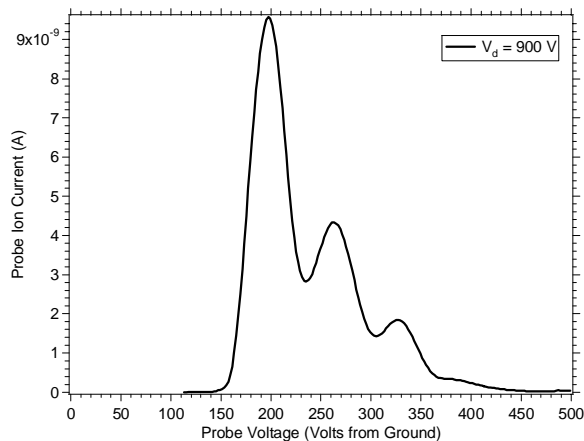


Figure 7.—ExB probe ion current versus probe voltage at 900 V.

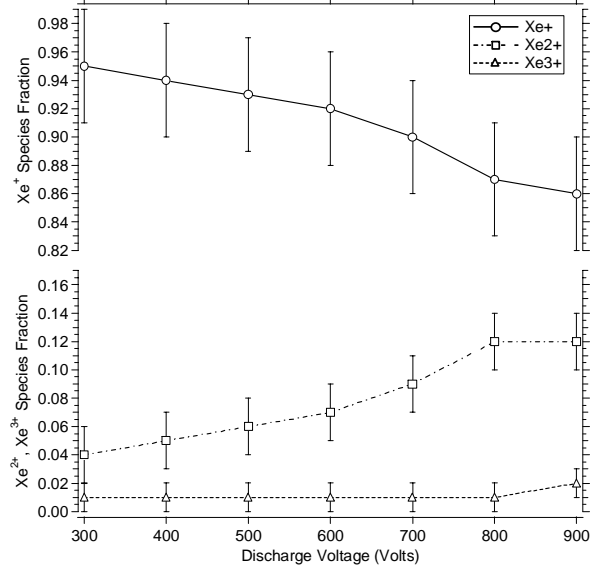


Figure 8.—Ion species fractions versus discharge voltage.

The variation of the ion species fractions with discharge voltage is shown in Figure 8 and tabulated in the appendix as Table 2. While the results indicated a general increase in the fraction of multiply-charged ions with increasing discharge voltage, Xe^+ still dominated the plasma at all discharge voltages. Over the range of 300 to 900 V, the species fraction of Xe^+ decreased from 0.95 to 0.86. Over the same voltage range, the Xe^{2+} species fractions increased from 0.04 to 0.12 and the Xe^{3+} species fraction increased from 0.01 to 0.02. At 300 V, the fractions of Xe^{2+} and Xe^{3+} measured by the ExB probe were lower, but still consistent with, previous Hall thruster studies using other mass analyzers.^{19,21,23–26} These past studies have shown the plume to be composed of 0.06 to 0.11 Xe^{2+} and 0.01 Xe^{3+} .

Multiply-charged ions reduce the thruster efficiency because

$$\frac{\eta}{\eta^+} = \frac{\left(\sum \Omega_{i,\text{beam}} / \sqrt{q_i} \right)^2}{\sum \Omega_{i,\text{beam}} / q_i} \quad (15)$$

where η^+ is the efficiency of a singly-charged plasma and $\Omega_{i,\text{beam}}$ are the beam current fractions. Note that the beam current fractions are different than the probe current fractions defined in Eq. (14)

because of the effects of secondary electron emission on the collected ion current. The right hand side of Eq. (15) is always less than one, so the efficiency of a multiply-charged plasma (for the same beam current), is always less than that of a singly-charged plasma. Applying Eq. (15) to the measured species fractions showed that the thruster efficiency decreased by 1 to 3% over 300 to 900 V due to the presence of multiply-charged ion species. While this is a small decrease in efficiency, Eq. (15) does not include the extra energy losses of producing multiply-charged ions, thermal losses due to ion acceleration into the discharge chamber walls, and the angular distribution of multiply-charged ions (measurements from thruster centerline usually show higher fractions of multiply-charged species).^{19,21,23–26} These effects, as well as the effects of multiply-charged ions on thruster lifetime, are currently being considered in further detail.

REFERENCES

1. Noca, M., Brophy, J.R., "Over Powering Solar System Exploration," AIAA-1997-2914, 33rd Joint Propulsion Conference, Seattle, WA, July 6–9, 1997.
2. Oleson, S.R., "Advanced Electric Propulsion for Space Solar Power Satellites," AIAA-1999-2872, 35th Joint Propulsion Conference, Los Angeles, CA, June 20–24, 1999.
3. Oleson, S.R., "Mission Advantages of Constant Power, Variable Isp Electrostatic Thrusters," AIAA-2000-3413, 36th Joint Propulsion Conference, Huntsville, AL, July 17–19, 2000.
4. Fiehler, D.I., Oleson, S.R., "A Comparison of Electric Propulsion Systems for Mars Exploration," AIAA-2003-4574, 39th Joint Propulsion Conference, Huntsville, AL July 20–23, 2003.
5. Hofer, R.R., Peterson, P.Y., Gallimore, A.D., "A High Specific Impulse Two-Stage Hall Thruster with Plasma Lens Focusing," IEPC-01-036, 27th International Electric Propulsion Conference, Pasadena, CA, Oct. 14–19, 2001.
6. Hofer, R.R., Gallimore, A.D., "The Role of Magnetic Field Topography in Improving the Performance of High-Voltage Hall Thrusters," AIAA-2002-4111, 38th Joint Propulsion Conference, Indianapolis, IN, July 7–10, 2002.
7. Hofer, R.R., Jankovsky, R.S., "The Influence of Current Density and Magnetic Field Topography in Optimizing the Performance, Divergence, and Plasma Oscillations of High Specific Impulse Hall Thrusters," IEPC-03-142, 28th International Electric Propulsion Conference, Toulouse, France, March 17–21, 2003.
8. Hofer, R.R., Gallimore, A.D., "Recent Results from Internal and Very-Near-Field Plasma Diagnostics of a High Specific Impulse Hall Thruster," IEPC-03-037, 28th International Electric Propulsion Conference, Toulouse, France, March 17–21, 2003.
9. Hofer, R.R., Gallimore, A.D., "Ion voltage diagnostics in the far-field plume of a high-specific impulse Hall thruster," AIAA-2003-4556, 39th Joint Propulsion Conference, Huntsville, AL July 20–23, 2003.
10. Seliger, R.L., "ExB Mass-Separator Design," Journal of Applied Physics, Vol. 43, No. 5, May 1972.
11. Vahrenkamp, R.P., "Measurement of Double Charged Ions in the Beam of a 30-cm Mercury Bombardment Thruster," AIAA-73-1057, AIAA 10th Electric Propulsion Conference, Lake Tahoe, NV, Oct. 31–Nov. 2, 1973.
12. Quiros, E.L., "Some Experimental Facts that Indicate the Elimination of Astigmatism in Ion Beams with Separators Using Crossed Electric and Magnetic Fields," Journal of Applied Physics, Vol. 52, No. 3, March 1981.
13. Batson, P.E., "High-energy resolution electron spectrometer for 1-nm spatial analysis," Review of Scientific Instruments, Vol. 57, No. 1, Jan. 1986.
14. Leal-Quiros, E., Prelas, M.A., "New Tilted-Poles Wien Filter with Enhanced Performance," Review of Scientific Instruments, Vol. 60, No. 3, March 1989.
15. Leal-Quiros, E., Prelas, M.A., "Focused High-Intensity Proton Beam from a Lithium Source by Using an ExB Stigmatic Selector," Review of Scientific Instruments, Vol. 61, No. 1, Jan. 1990.

16. Guan, S., Marshal, A.G., "A Mass- and Velocity-Broadband Ion Deflector for Off-Axis Ion Injection Into a Cyclotron Resonance Ion Trap," *Review of Scientific Instruments*, Vol. 67, No. 2, Feb. 1996.
17. Anderson, J.R., Fitzgerald, D., "Fullerene Propellant Research for Electric Propulsion," AIAA-96-3211, 1996.
18. Wrenger, B., Meiwes-Broer, K.H., "The Application of a Wien Filter to Mass Analysis of Heavy Clusters from a Pulsed Supersonic Nozzle Source," *Review of Scientific Instruments*, Vol. 68, No. 5, May 1997.
19. Kim, S.W., "Experimental Investigations of Plasma Parameters and Species-Dependent Ion Energy Distribution in the Plasma Exhaust Plume of a Hall Thruster," Ph.D. Dissertation, University of Michigan, 1999.
20. Williams, G.J., Domonkos, M.T., Chavez, J.M., "Measurement of Doubly Charged Ions in Ion Thruster Plumes," IEPC-2001-310, 27th International Electric Propulsion Conference, Pasadena, CA, Oct. 14-19, 2001.
21. Kim, S.W., Gallimore, A.D., "Plume Study of a 1.35-kW SPT-100 Using an ExB Probe," *Journal of Spacecraft and Rockets*, Vol. 39, No. 6, Nov.-Dec. 2002.
22. Hofer, R.R., Jankovsky, R.S., "A Hall Thruster Performance Model Incorporating the Effects of a Multiply-Charged Plasma," AIAA-2001-3322, 37th Joint Propulsion Conference, Salt Lake City, UT, July 8-11, 2001.
23. King, L.B., "Transport-property and Mass Spectral Measurements in the Plasma Exhaust Plume of a Hall-effect Space Propulsion System," Ph.D. Dissertation, University of Michigan, 1998.
24. Gulczinski, F.S., "Examination of the Structure and Evolution of Ion Energy Properties of a 5 kW Class Laboratory Hall Effect Thruster at Various Operational Conditions," Ph.D. Dissertation, University of Michigan, 1999.
25. King, L.B., Gallimore, A.D., "Mass Spectral Measurements in the Plume of an SPT-100 Hall Thruster," *Journal of Propulsion and Power*, Vol. 16, No. 6, Nov.-Dec. 2000.
26. Gulczinski, F.S., Gallimore, A.D., "Near-Field Ion Energy and Species Measurements of a 5-kW Hall Thruster," *Journal of Propulsion and Power*, Vol. 17, No. 2, March-April 2001.
27. Hagstrum, H.D., "Auger Ejection of Electrons from Tungsten by Noble Gas Ion," *Physical Review*, Vol. 96, No. 2, pp. 325-335, 1954.

APPENDIX

Table 1.—NASA-173Mv2 telemetry from the ExB probe measurements.

Point	Vd (V)	Id (A)	Anode (mg/s)	Cathode (mg/s)	Inner Coil (A)	Outer Coil (A)	Internal Trim Coil (A)	External Trim Coil (A)	Vcg (V)	Pressure (Torr)
93	300.4	8.87	10.00	1.00	1.75	1.50	0.00	0.00	-11.8	4.6E-06
101	400.4	9.24	10.00	1.00	2.50	2.26	0.00	0.00	-13.0	4.6E-06
105	500.4	9.37	10.00	1.00	3.00	2.20	0.00	0.00	-13.3	4.6E-06
109	600.2	9.53	10.00	1.00	3.26	2.49	0.00	0.00	-13.4	4.6E-06
113	700.4	9.71	10.00	1.00	3.41	3.06	0.00	0.00	-14.0	4.6E-06
117	800.1	9.81	10.00	1.00	3.82	3.30	0.00	0.00	-14.1	4.6E-06
126	800.1	9.84	10.00	1.00	3.85	3.61	0.00	0.00	-13.8	4.6E-06

Table 2.—Ion species fractions on centerline, two meters downstream of the NASA-173Mv2 at 10 mg/s.

Vd	Species Fraction		
	Xe+	Xe2+	Xe3+
300	0.95	0.04	0.01
400	0.94	0.05	0.01
500	0.93	0.06	0.01
600	0.92	0.07	0.01
700	0.90	0.09	0.01
800	0.87	0.12	0.01
900	0.86	0.12	0.02

REPORT DOCUMENTATION PAGE			Form Approved OMB No. 0704-0188	
Public reporting burden for this collection of information is estimated to average 1 hour per response, including the time for reviewing instructions, searching existing data sources, gathering and maintaining the data needed, and completing and reviewing the collection of information. Send comments regarding this burden estimate or any other aspect of this collection of information, including suggestions for reducing this burden, to Washington Headquarters Services, Directorate for Information Operations and Reports, 1215 Jefferson Davis Highway, Suite 1204, Arlington, VA 22202-4302, and to the Office of Management and Budget, Paperwork Reduction Project (0704-0188), Washington, DC 20503.				
1. AGENCY USE ONLY (Leave blank)		2. REPORT DATE December 2003		3. REPORT TYPE AND DATES COVERED Final Contractor Report
4. TITLE AND SUBTITLE Ion Species Fractions in the Far-Field Plume of a High-Specific Impulse Hall Thruster			5. FUNDING NUMBERS WBS-22-755-04-12 NAS3-00145	
6. AUTHOR(S) Richard R. Hofer and Alec D. Gallimore				
7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES) QSS Group, Inc. 21000 Brookpark Road Cleveland, Ohio 44135			8. PERFORMING ORGANIZATION REPORT NUMBER E-14311	
9. SPONSORING/MONITORING AGENCY NAME(S) AND ADDRESS(ES) National Aeronautics and Space Administration Washington, DC 20546-0001			10. SPONSORING/MONITORING AGENCY REPORT NUMBER NASA CR-2003-212893 AIAA-2003-5001	
11. SUPPLEMENTARY NOTES Prepared for the 39th Joint Propulsion Conference and Exhibit cosponsored by AIAA, ASME, SAE, and ASEE, Huntsville, Alabama, July 20-23, 2003. Richard R. Hofer, QSS Group, Inc., Cleveland, Ohio 44135; and Alec D. Gallimore, University of Michigan, Plasmadynamics and Electric Propulsion Laboratory, Ann Arbor, Michigan 48109. Project Manager, David Jacobson, Power and On-Board Propulsion Technology Division, NASA Glenn Research Center, organization code 5430, 216-433-3691.				
12a. DISTRIBUTION/AVAILABILITY STATEMENT Unclassified - Unlimited Subject Category: 20 Available electronically at http://gltrs.grc.nasa.gov This publication is available from the NASA Center for AeroSpace Information, 301-621-0390.			12b. DISTRIBUTION CODE	
13. ABSTRACT (Maximum 200 words) An ExB probe was used to measure the ion species fractions of Xe^+ , Xe^{2+} , and Xe^{3+} in the far-field plume of the NASA-173Mv2 laboratory-model Hall thruster. The thruster was operated at a constant xenon flow rate of 10 mg/s and discharge voltages of 300 to 900 V. The ExB probe was placed two meters downstream of the thruster exit plane on the thruster centerline. At a discharge voltage of 300 V, the species fractions of Xe^{2+} and Xe^{3+} were lower, but still consistent with, previous Hall thruster studies using other mass analyzers. Over discharge voltages of 300 to 900 V, the Xe^{2+} species fractions increased from 0.04 to 0.12 and the Xe^{3+} species fraction increased from 0.01 to 0.02.				
14. SUBJECT TERMS Hall thruster			15. NUMBER OF PAGES 16	
			16. PRICE CODE	
17. SECURITY CLASSIFICATION OF REPORT Unclassified	18. SECURITY CLASSIFICATION OF THIS PAGE Unclassified	19. SECURITY CLASSIFICATION OF ABSTRACT Unclassified	20. LIMITATION OF ABSTRACT	