National Aeronautics and Space Administration

Heavy Lift for National Security: The Ares V

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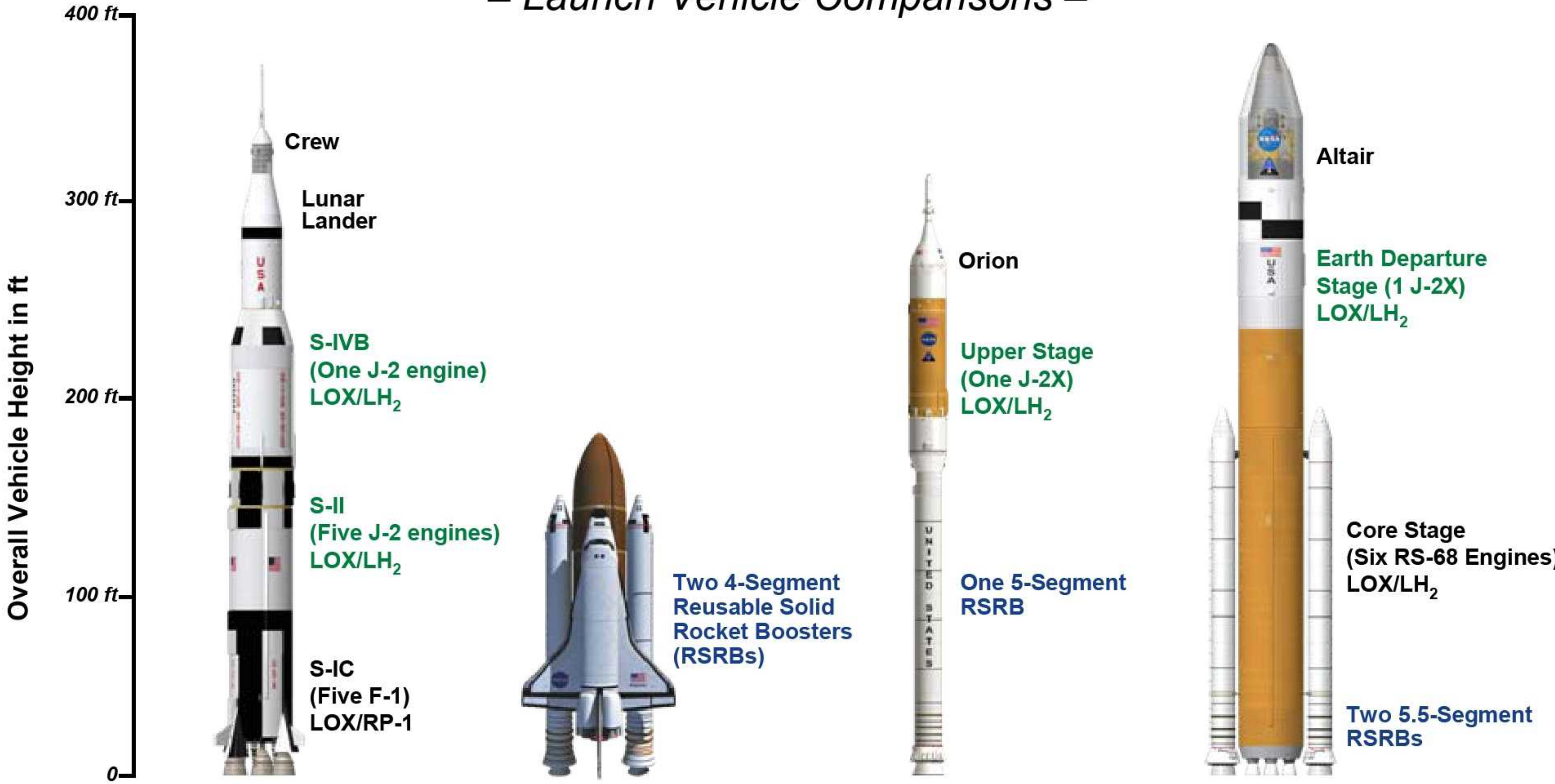
Introduction

- The NASA Ares Projects Office is developing the launch vehicles to move the United States and humanity beyond low earth orbit
- Ares I is a crewed vehicle, and Ares V is a heavy lift vehicle being designed to launch cargo into LEO and transfer cargo and crews to the Moon
- This is a snapshot of development and capabilities. Ares V is early in the requirements formulation stage of development pending the outcome of the Review of U.S. Human Space Flight Plans Committee and White House action.
- The Ares V vehicle will be considered a national asset, creating unmatched opportunities for human exploration, science, national security, and space business

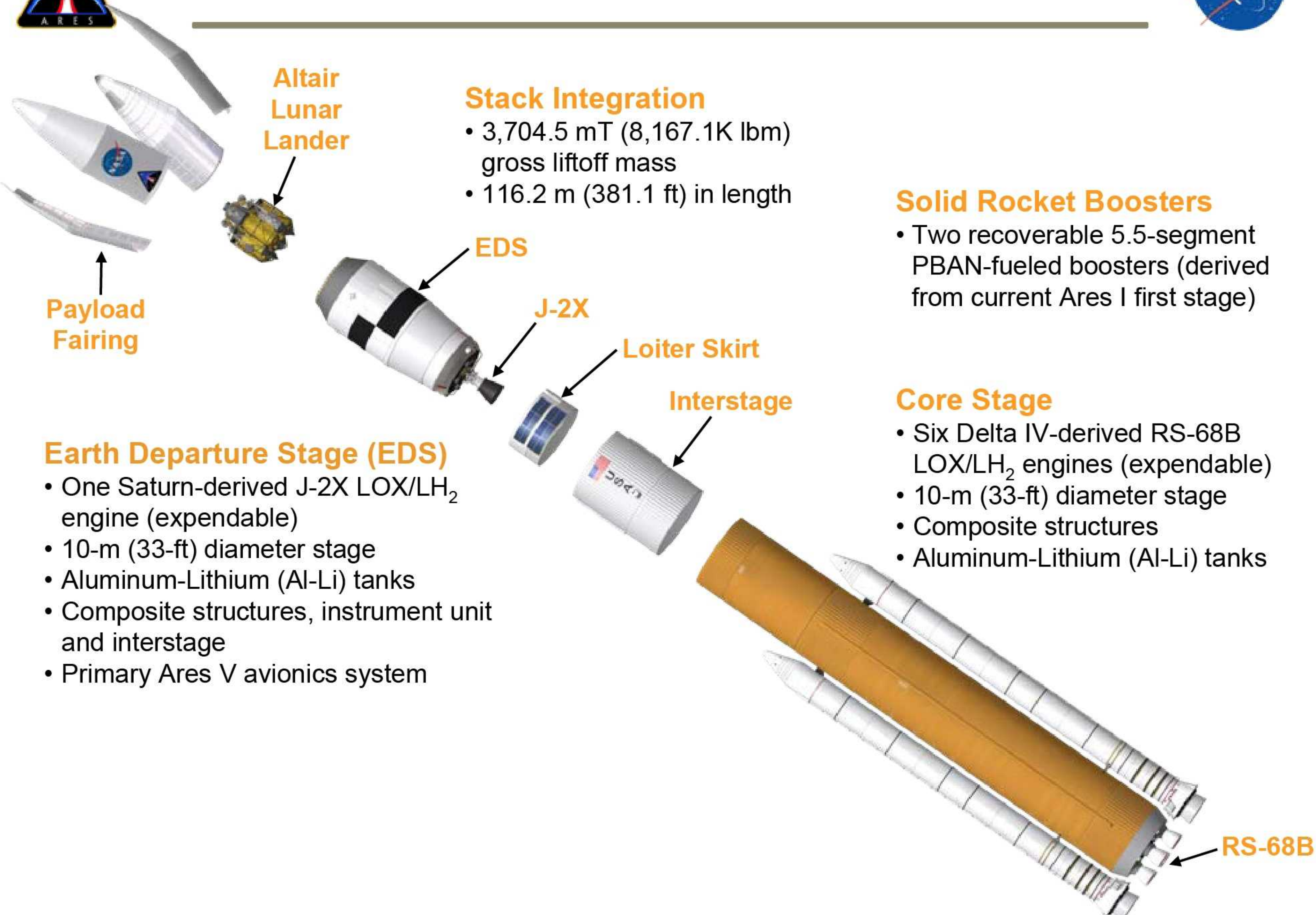
Building on 50 Years of Proven Experience

– Launch Vehicle Comparisons –



	Saturn V: 1967–1972	**Space Shuttle:** 1981–Present	**Ares I:** First Flight 2015	**Ares V:** First Flight 2018
Height	360 ft	184.2 ft	325.0 ft	381.1 ft
Gross Liftoff Mass (GLOM)	2,948.4 mT (6,500K lbm)	2,041.1 mT (4,500.0K lbm)	933.2 mT (2,057.3K lbm)	3,704.5 mT (8,167.1K lbm)
Payload Capability	99.0K lbm to TLI 262.0K lbm to LEO	55.1K lbm to LEO	54.9K lbm to LEO	156.7K lbm to TLI with Ares I 413.8K lbm to LEO

Ares V Elements



Stack Integration

- 3,704.5 mT (8,167.1K lbm) gross liftoff mass
- 116.2 m (381.1 ft) in length

Solid Rocket Boosters

- Two recoverable 5.5-segment PBAN-fueled boosters (derived from current Ares I first stage)

Earth Departure Stage (EDS)

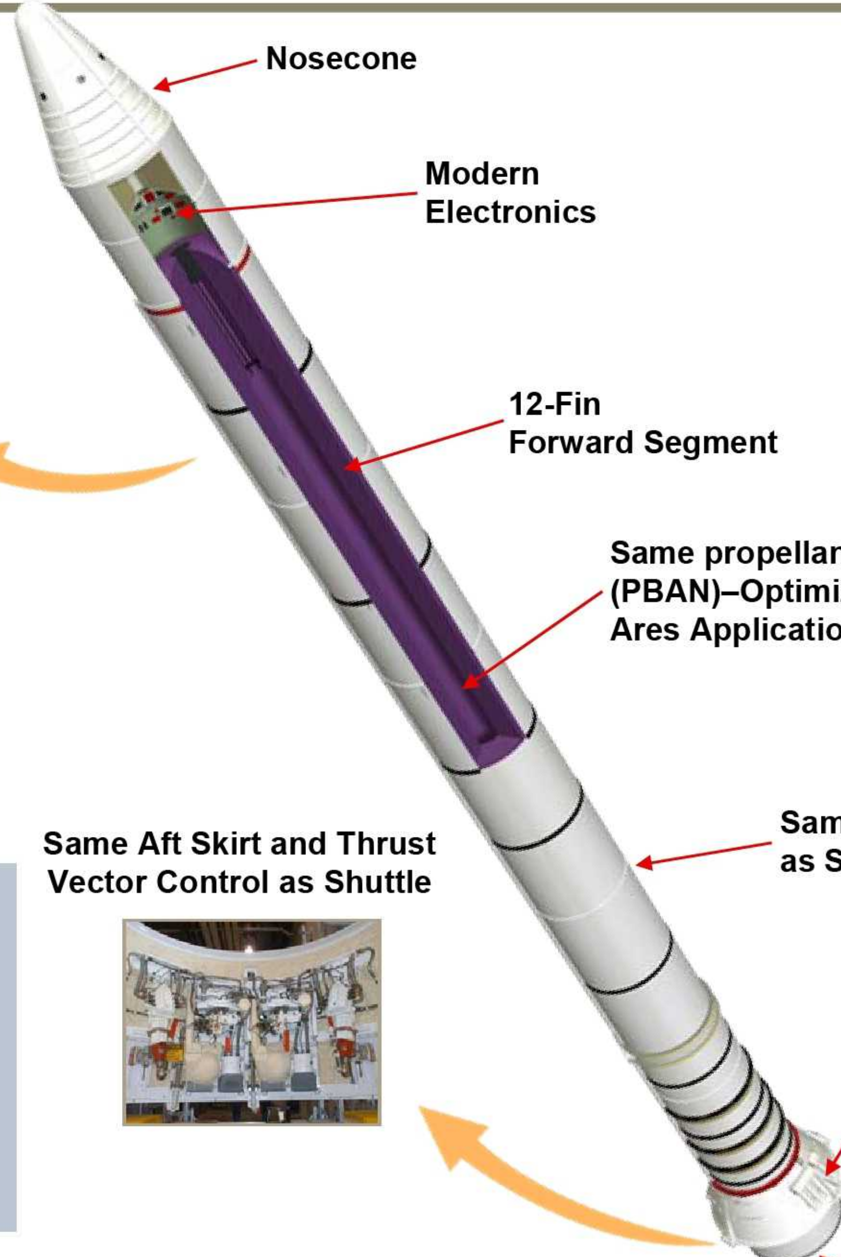
- One Saturn-derived J-2X LOX/LH_2 engine (expendable)
- 10-m (33-ft) diameter stage
- Aluminum-Lithium (Al-Li) tanks
- Composite structures, instrument unit and interstage
- Primary Ares V avionics system

Core Stage

- Six Delta IV-derived RS-68B LOX/LH_2 engines (expendable)
- 10-m (33-ft) diameter stage
- Composite structures
- Aluminum-Lithium (Al-Li) tanks

Ares V Solid Rocket Booster (SRB)

Ares V SRB is similar to Space Shuttle and Ares I but optimized for lunar missions



Same Aft Skirt and Thrust Vector Control as Shuttle

Each Booster:

Mass: 791.5 t (1,744.9 klb_m)

Thrust: 16.86 MN (3.79 Mlb_f)

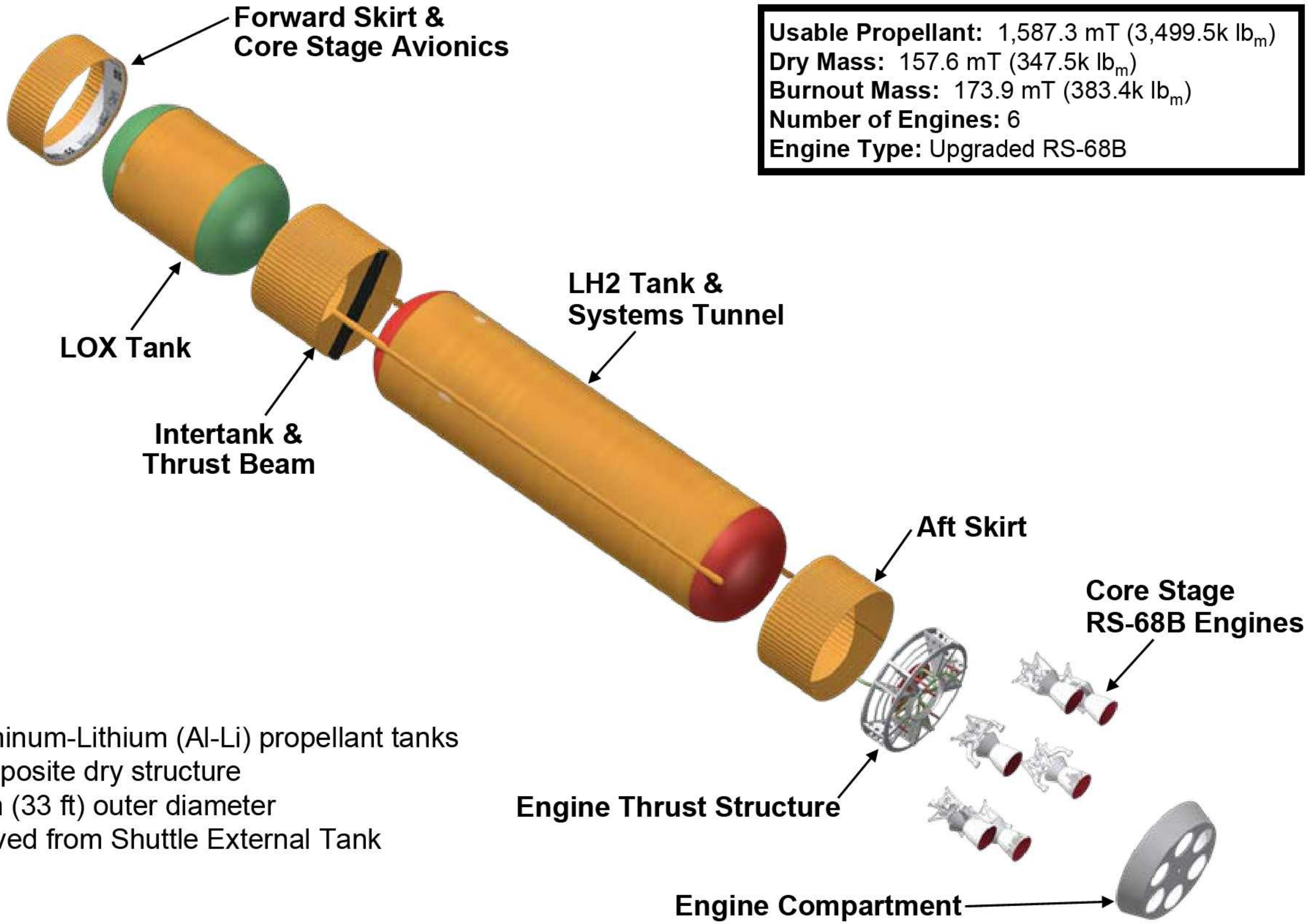
Burn Duration: 126 sec

Height: 59 m (193 ft)

Diameter: 3.7 m (12 ft)

Ares V Core Stage

Usable Propellant:	1,587.3 mT (3,499.5k lb_m)
Dry Mass:	157.6 mT (347.5k lb_m)
Burnout Mass:	173.9 mT (383.4k lb_m)
Number of Engines:	6
Engine Type:	Upgraded RS-68B

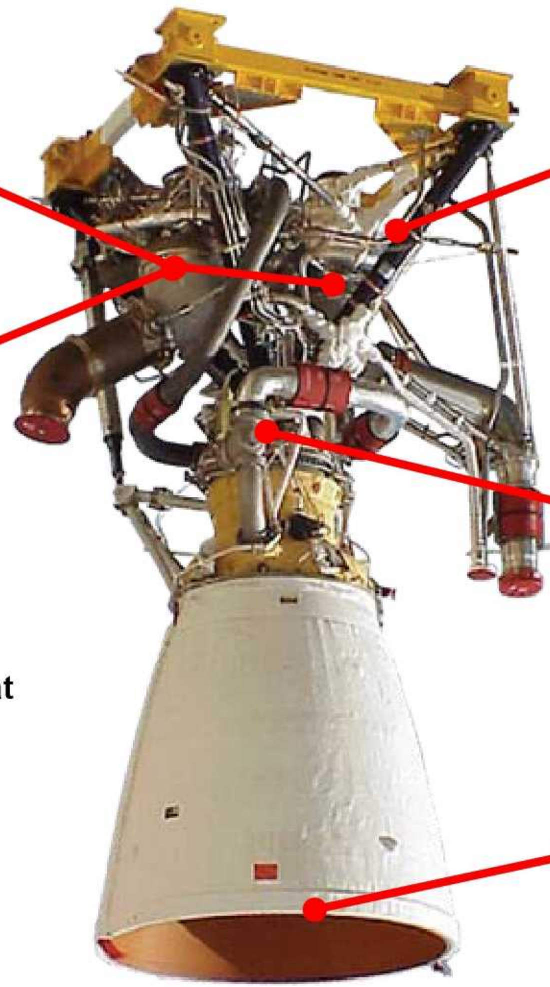


- Aluminum-Lithium (Al-Li) propellant tanks
- Composite dry structure
- 10 m (33 ft) outer diameter
- Derived from Shuttle External Tank

RS-68 to RS-68B

* Redesigned turbine nozzles to increase maximum power level by ≈ 2%

Redesigned turbine seals to significantly reduce helium usage for pre-launch



Helium spin-start duct redesign, along with start sequence modifications, to help minimize pre-ignition free hydrogen

* Higher element density main injector improving specific impulse by ≈ 2% and thrust by ≈ 4%

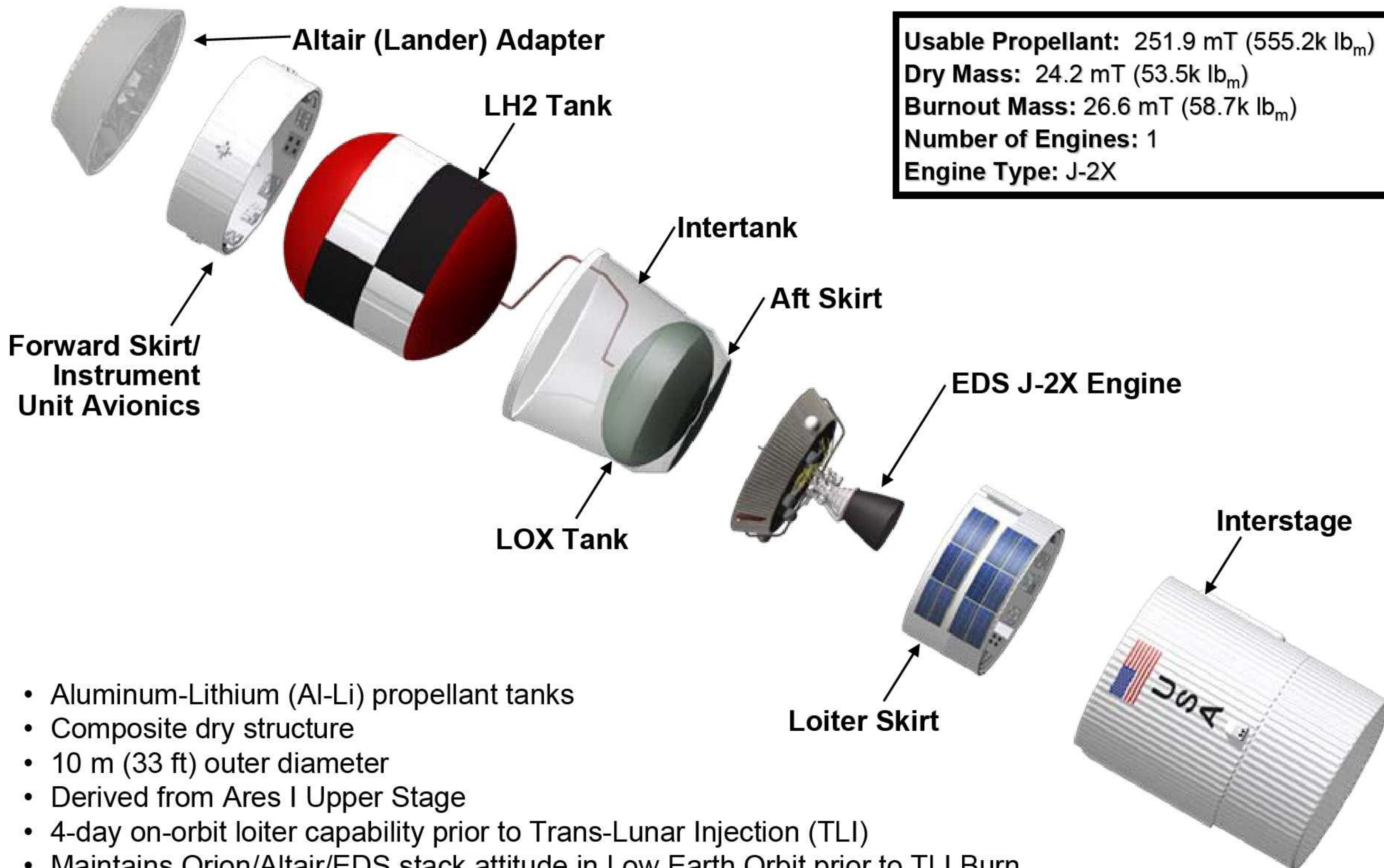
Increased duration capability ablative nozzle

- **Other RS-68A upgrades or changes that may be included:**
  - Bearing material change
  - New Gas Generator igniter design
  - Improved Oxidizer Turbo Pump temp sensor
  - Improved hot gas sensor
  - 2nd stage Fuel Turbo Pump blisk crack mitigation
  - Cavitation suppression
  - ECU parts upgrade

* **RS-68A Upgrades**

Ares V Earth Departure Stage

Usable Propellant: 251.9 mT (555.2k lb_m)
Dry Mass: 24.2 mT (53.5k lb_m)
Burnout Mass: 26.6 mT (58.7k lb_m)
Number of Engines: 1
Engine Type: J-2X



- Aluminum-Lithium (Al-Li) propellant tanks
- Composite dry structure
- 10 m (33 ft) outer diameter
- Derived from Ares I Upper Stage
- 4-day on-orbit loiter capability prior to Trans-Lunar Injection (TLI)
- Maintains Orion/Altair/EDS stack attitude in Low Earth Orbit prior to TLI Burn
- EDS provide 1.5 kW of power to Altair from launch to TLI

J-2X Engine

Used on Ares I and Ares V

Turbomachinery
- Based on J-2S MK-29 design

Gas Generator
- Based on RS-68 design

Engine Controller
- Based directly on RS-68 design and software architecture

Regeneratively Cooled Nozzle Section
- Based on long history of RS-27 success

Flexible Inlet Ducts
- Based on J-2 & J-2S ducts

Open-Loop Pneumatic Control
- Similar to J-2

HIP-bonded MCC
- Based on RS-68 demonstrated technology

Metallic Nozzle Extension
- New design

Mass: 2.5 mT (5,511 lbm)

Height: 4.7 m (15.4 ft)

Diameter: 3.05 m (10 ft)

Thrust: 1,308K N (294K lbm) (vac)

Isp: 448 sec (vac)

Operation Time: 500 sec.

Altitude Start / On-orbit Restart

Operational Life: 8 starts/ 2,600 sec

Pratt & Whitney
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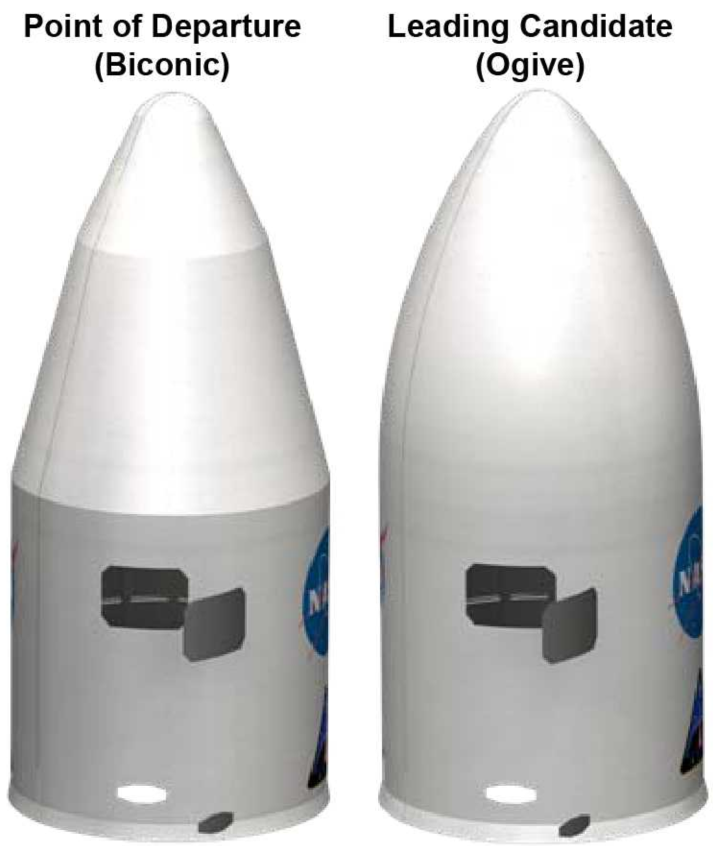
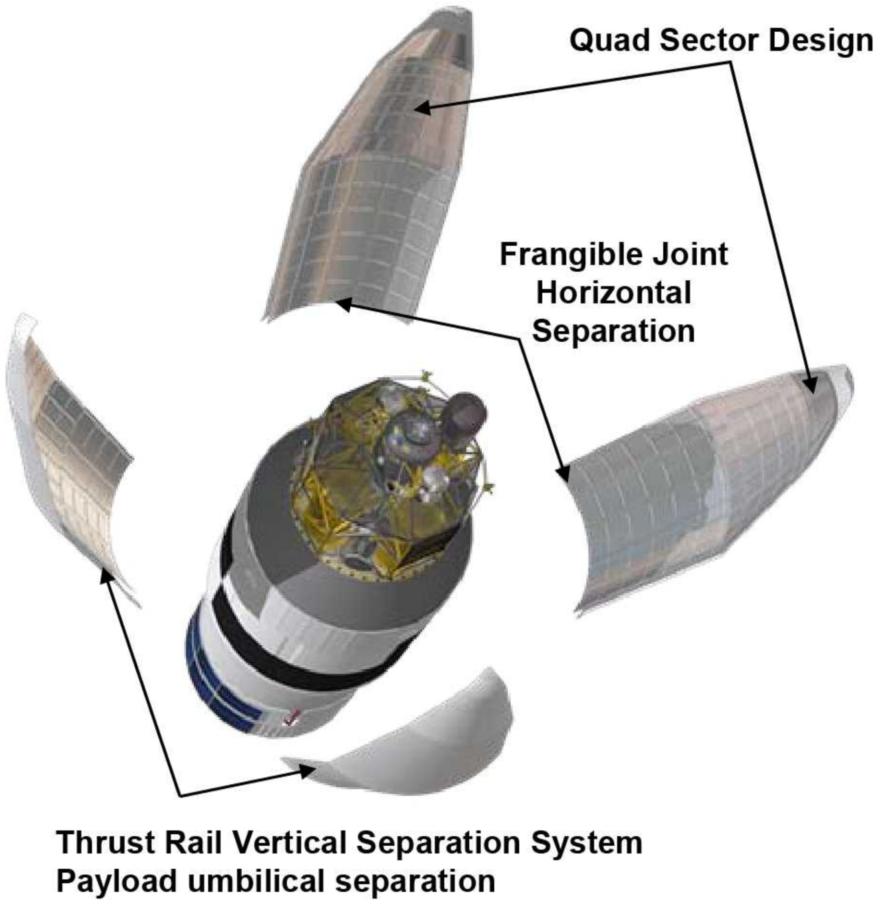
Payload Shroud Point Of Departure

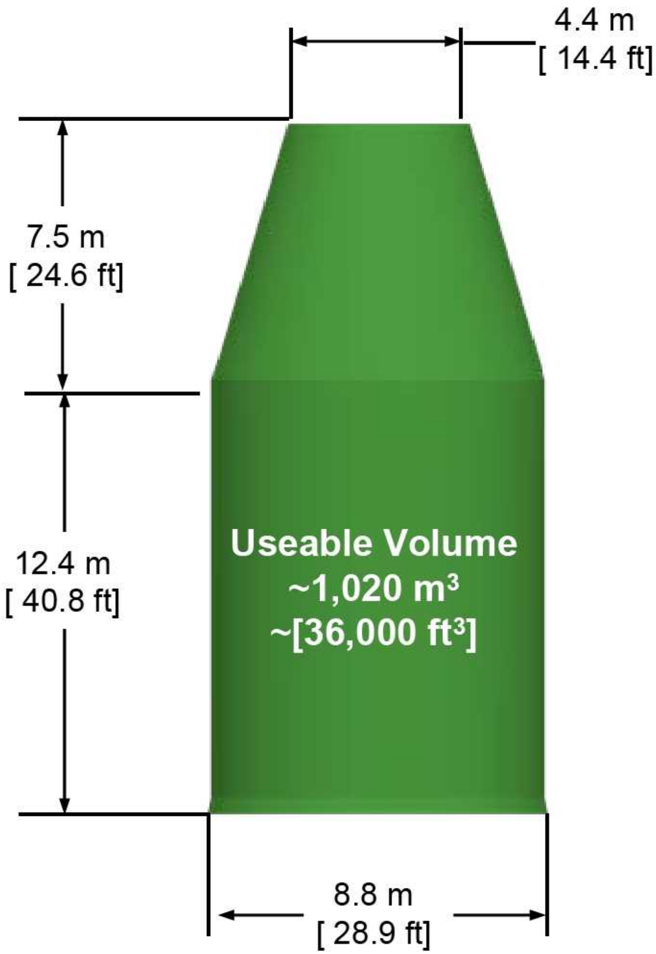


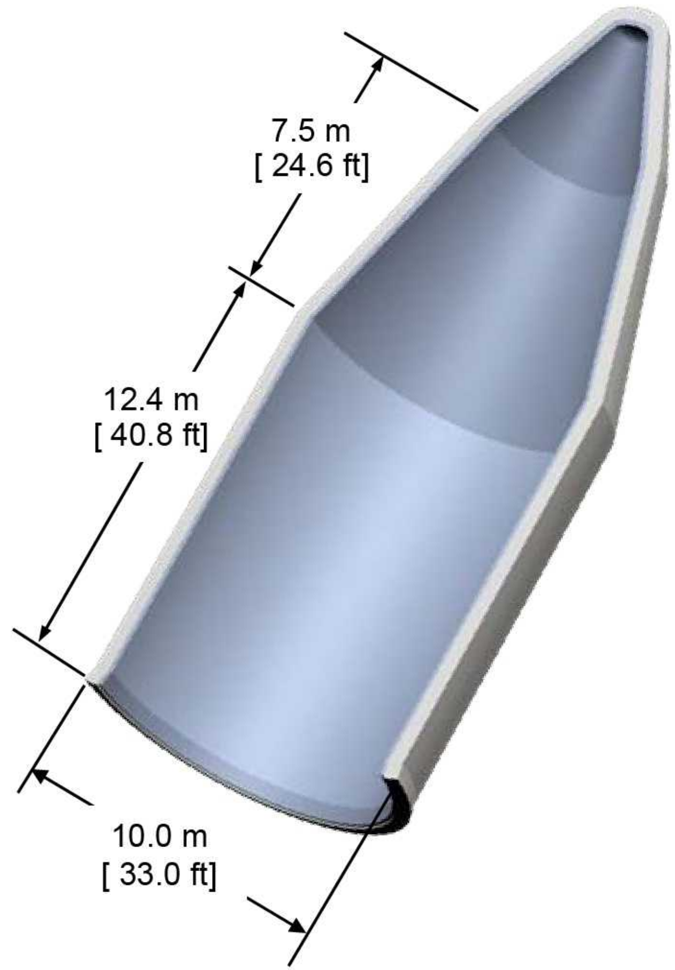
Mass: 9.1 mT (20.0k lbm)
POD Geometry: Biconic
Design: Quad sector
Barrel Diameter: 10 m (33 ft)
Barrel Length: 9.7 m (32 ft)
Total Length: 22 m (72ft)



- Composite sandwich construction (Carbon-Epoxy face sheets, Al honeycomb core)
- Painted cork TPS bonded to outer face sheet with RTV
- Payload access ports for maintenance, payload consumables and environmental control (while on ground)

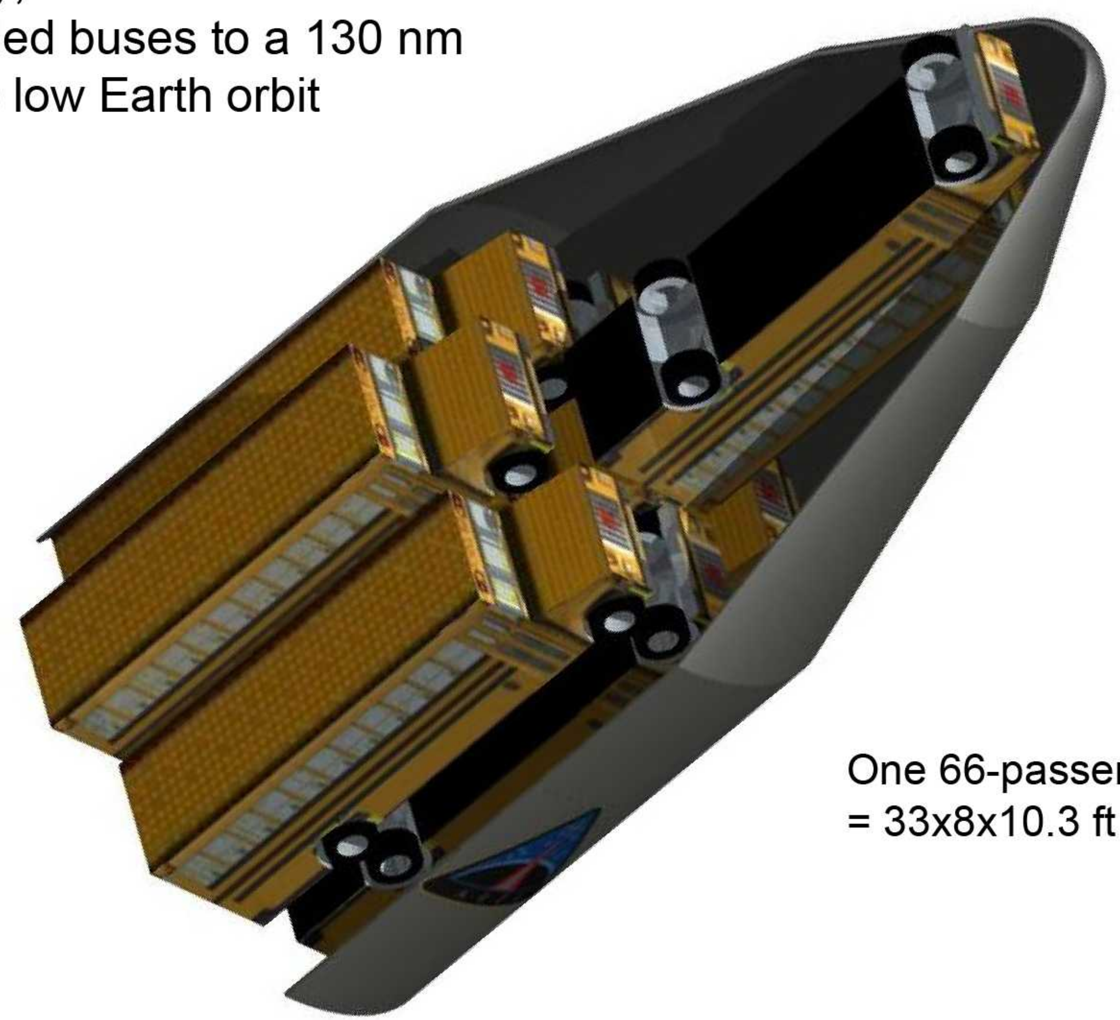
Current Ares V Shroud Concept





Visualizing the Possibilities

Combining mass and volume capability, Ares V could launch 8 fully loaded buses to a 130 nm (circular) low Earth orbit



One 66-passenger school bus
= 33x8x10.3 ft / 20,100 lb empty

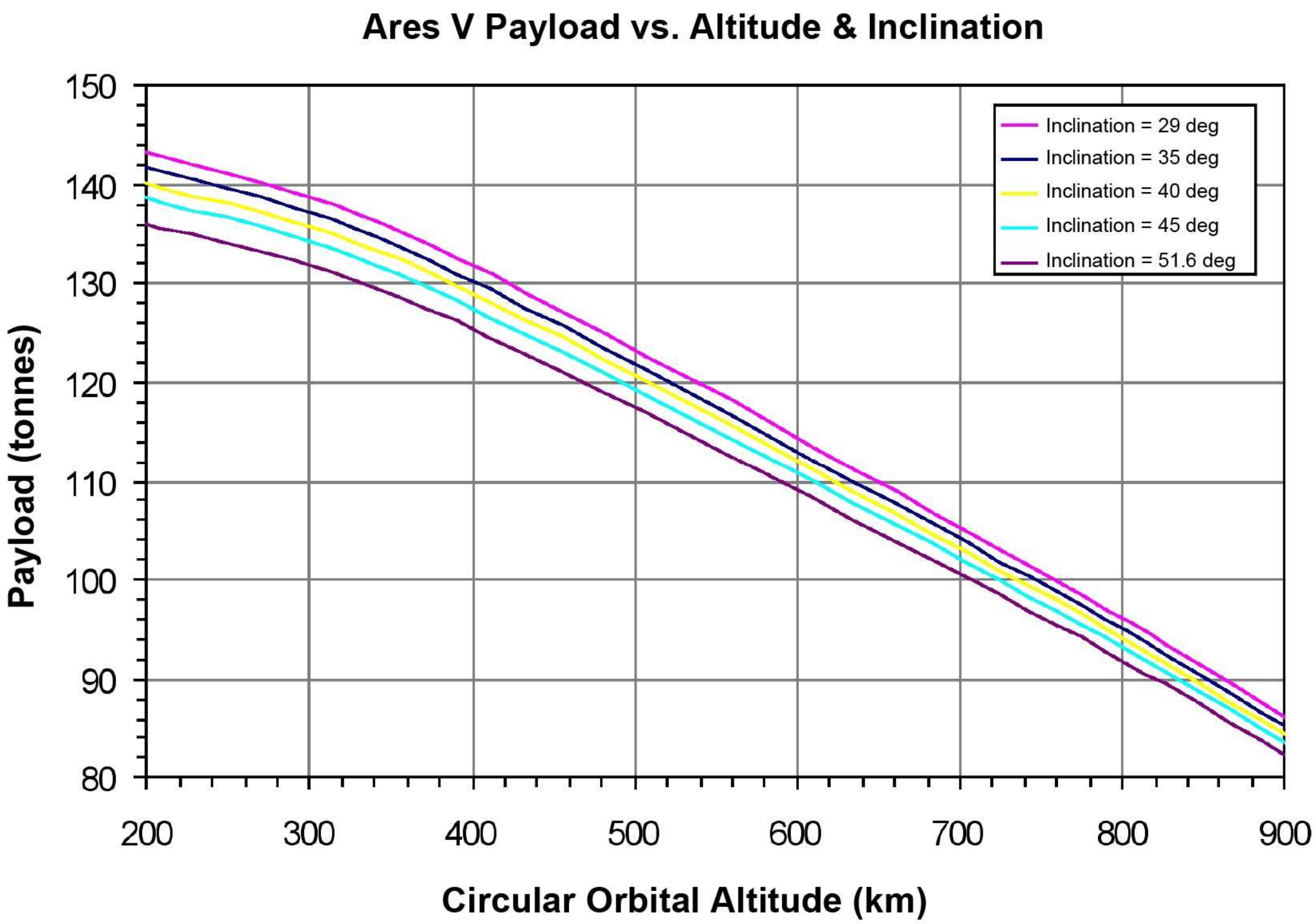
Ares V Performance for Selected Trajectories from KSC

Mission Profile	Target	Constellation POD Shroud		Extended Shroud	
		Payload (lbm)	Payload (mt)	Payload (lbm)	Payload (mt)
1) LEO (@29° inclination)	241 x 241 km	315,000	143	313,000	142
2) GEO	Transfer DV 14,100 ft/s	77,000	35	76,000	34.5
3) GTO Injection	Transfer DV 8,200 ft/s	153,000	69.5	152,000	69
4) Sun-Earth L2 Transfer Orbit Injection	C3 of -0.7 km^2/s^2	124.000	56.5	123,000	56
5) Earth-Moon L2 Transfer Orbit Injection	C3 of -1.7 km^2/s^2	126,000	57.0	125,000	57
6) Cargo Lunar Outpost (TLI Direct), Reference	C3 of -1.8 km^2/s^2	126,000	57	125,000	57
7) Mars Cargo (TMI Direct)	C3 of 9 km^2/s^2	106,000	48	105,000	48

*based on LV 51.00.39

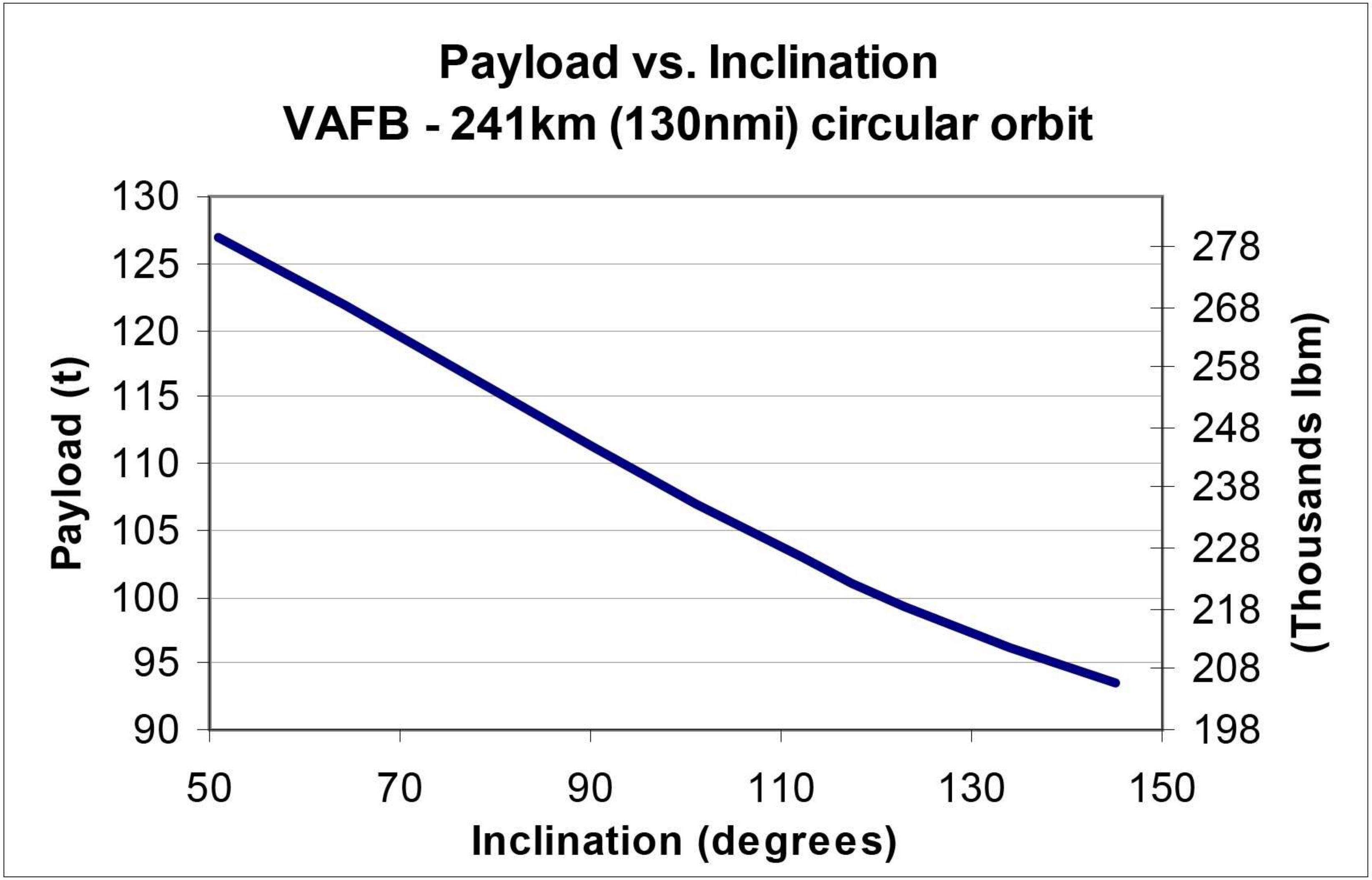
Ares V (51.00.39) LEO Performance

Previous POD Shroud



LEO performance for new Constellation point of departure vehicle (51.00.48) is expected to exceed values shown here. Performance analysis will be updated for the 51.00.48 vehicle.

Ares V LEO Capability from VAFB

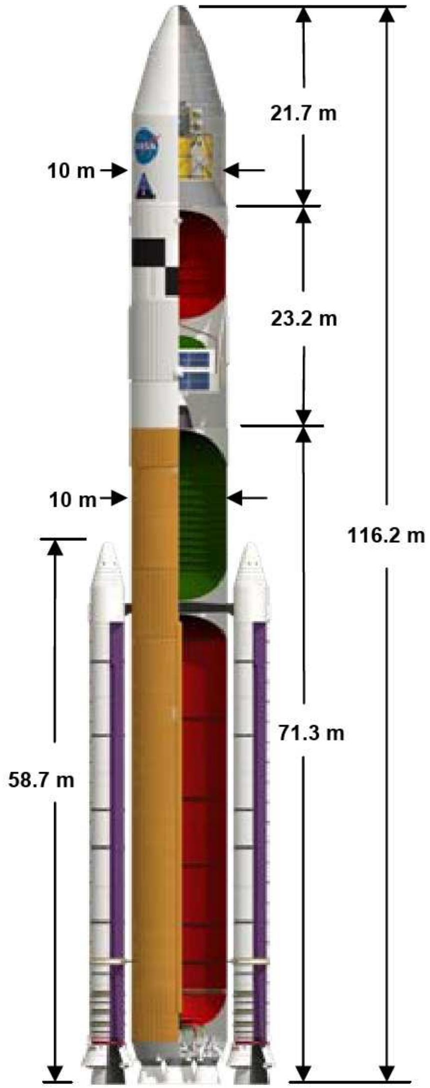


Approximate Performance – does not take into account land over-flight

December 2007 51.02.06 Configuration

LCCR/MCR-Approved Point of Departure

- Vehicle 51.0.48 -

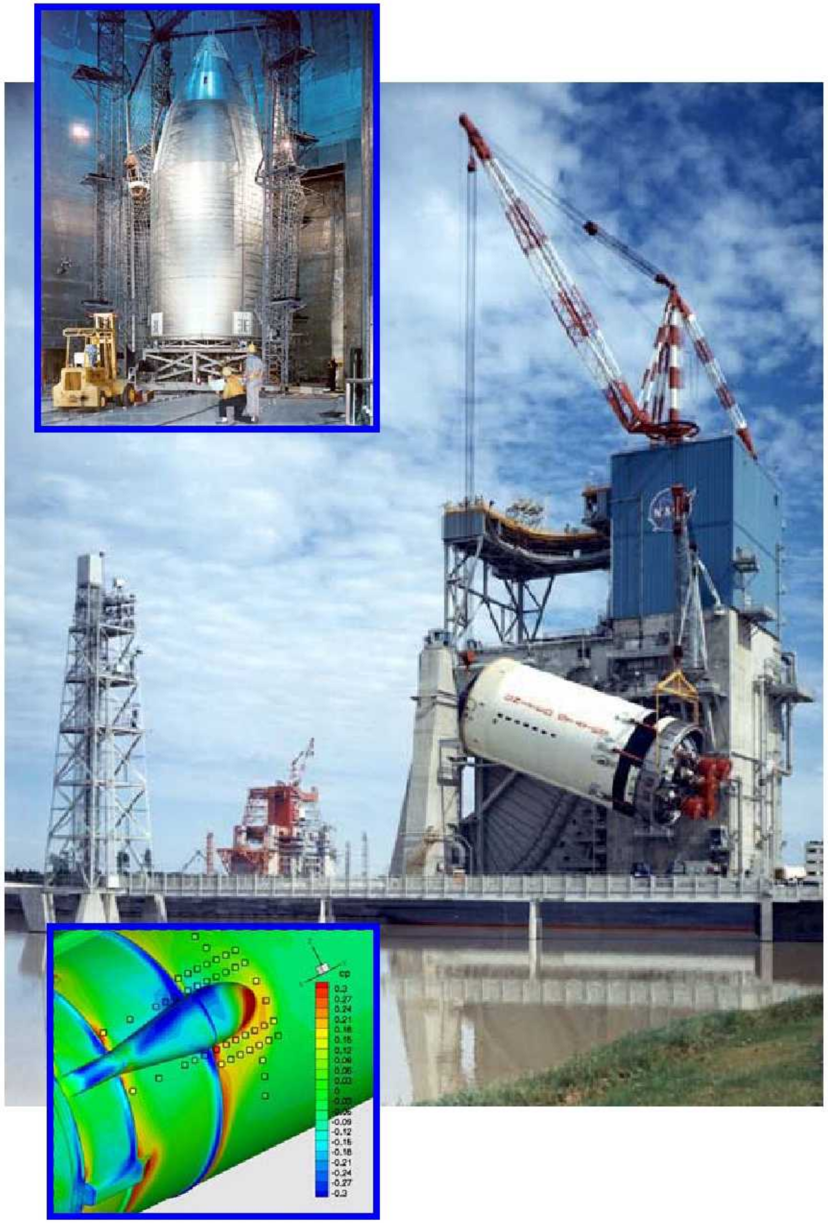


NOTE: These are MEAN numbers

- **Vehicle 51.0.48 approved in 2008**
  - 6 Engine Core, 5.5 Segment PBAN steel case booster
  - Provides architecture closure with margin
- **Approved maintaining Vehicle 51.0.47 with composite HTPB booster as Ares V option**
  - Final decision on Ares V booster at Constellation Lunar SRR (2010)
  - Additional performance capability if needed for margin or requirements
  - Allows for competitive acquisition environment for booster
- **Near Term Plan to Maintain Booster Options**
  - Fund key technology areas: composite cases, HTPB propellant characterization
  - Competitive Phase 1 industry concept definition proposals submitted early 2009.

Current Activities

- Ares V concept definition/requirements development industry proposals
- Structural test approach
- Structural test articles
- Ares V-Y flight test objectives
- Ares V aerodynamic characterization
- Manufacturing, test, and launch facilities
- Core Stage and EDS propulsion test approach and facilities assessment
- Technology prioritization
- Ares V Cost threat risk assessment
- Ares V performance risk assessment



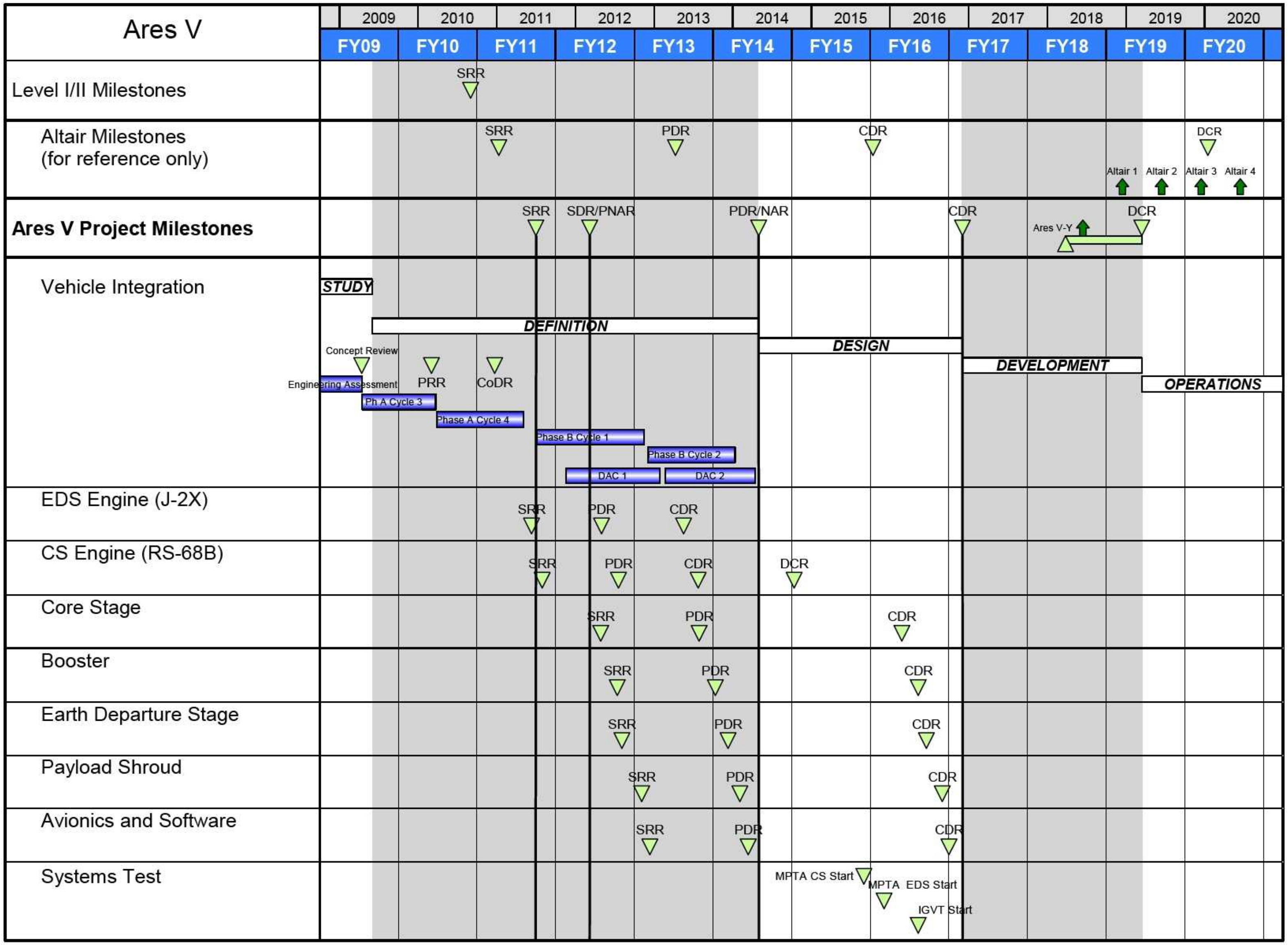
Conclusions

- Ares V current concept (51.00.48) exceeds Saturn V mass capability to trans lunar injection by almost 40% alone or almost 60% with Ares I
- This concept vehicle can meet current Human Lunar Return requirements with ~6 mT of margin
- 2009 activities focused on refining vehicle and operational concept, refining requirements, working with potential non-Constellation users to understand vehicle/payload benefits and design issues
- Ares V is sensitive to loiter time, attitude, power, and altitude requirements, in addition to payload performance
- Ares V is a national asset that can change today's rules for mass/volume or development cycle/cost.

www.nasa.gov/ares

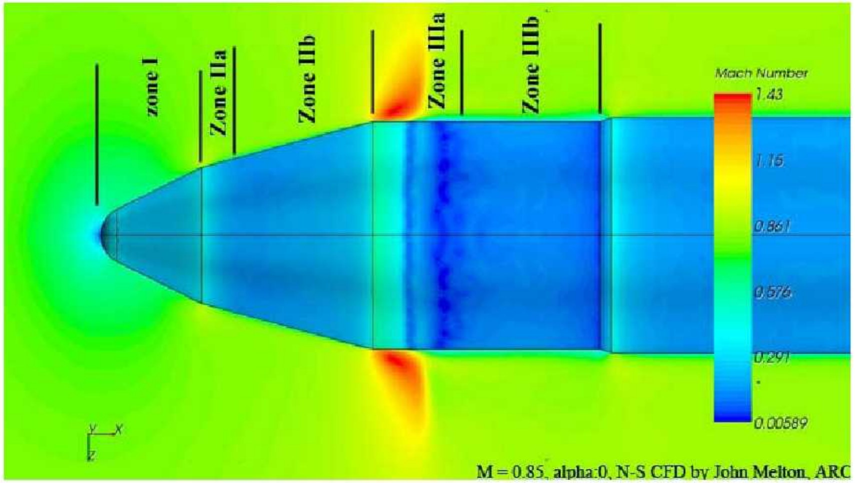
Backup

Ares V Summary Schedule



Preliminary Aero-acoustic Analysis

Transonic and Max-Q Acoustics



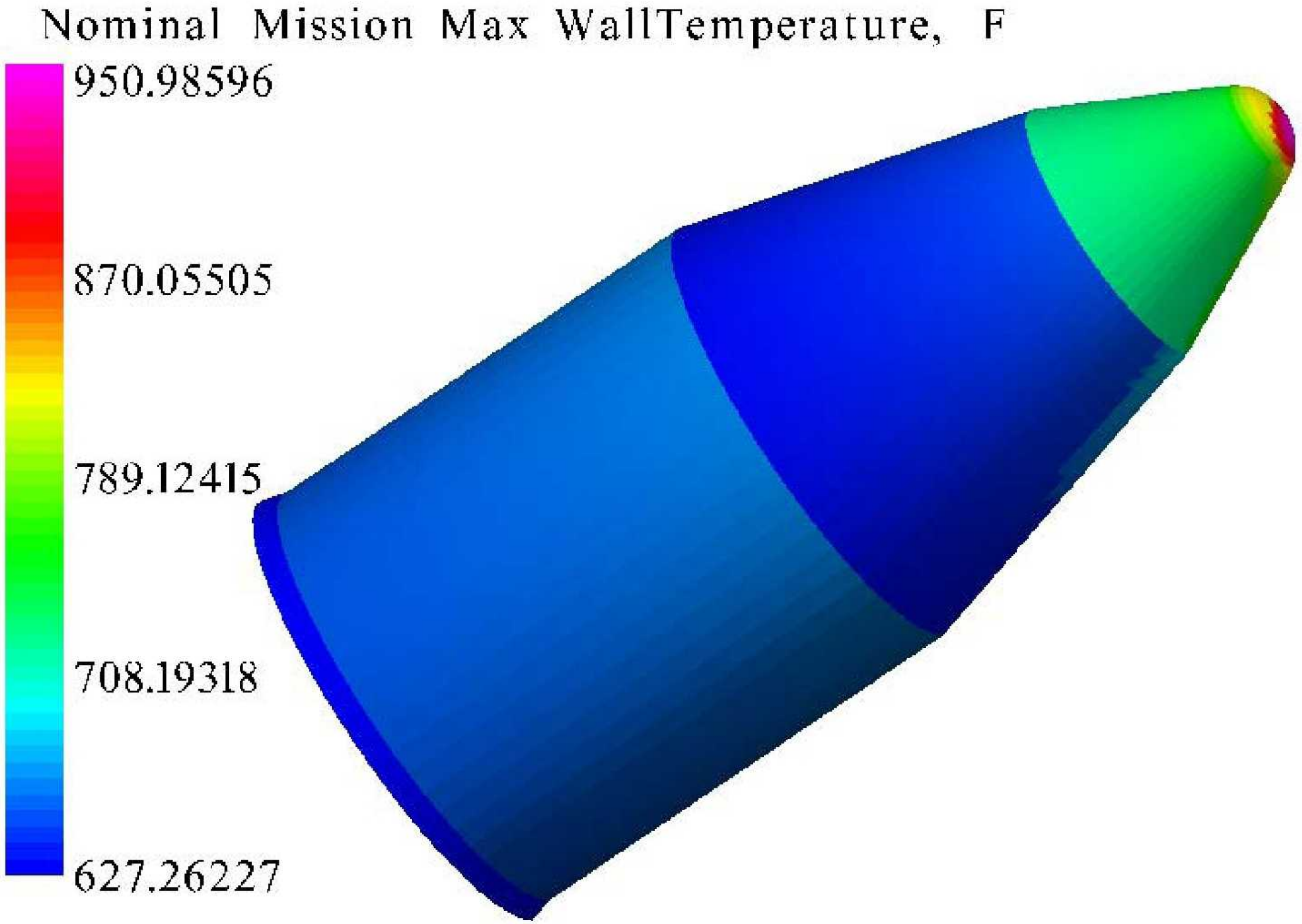
- **Predicted ascent max-acoustic levels**
- **Conceptual design based on acoustic blanket thicknesses used on Cassini mission**

Table I. Estimated max Overall Fluctuating Pressure Level (OAFPL) on Shroud external regions

Zone	I	IIa	IIb	IIIa	IIIb
Criteria for Max OAFPL	Attached Turbulent Boundary Layer	Weak Transonic Shock	Attached Turbulent Boundary Layer	Strong Transonic Shock & Separation	Weak Transonic Shock
Expected Mach # for max OAFPL	1.65	0.93	1.65	0.85	0.85
Q (psf)	707	520	707	475	475
Crms	0.007	0.07	0.007	0.12	0.035
OAFPL (dB)	142	159	142	163	152

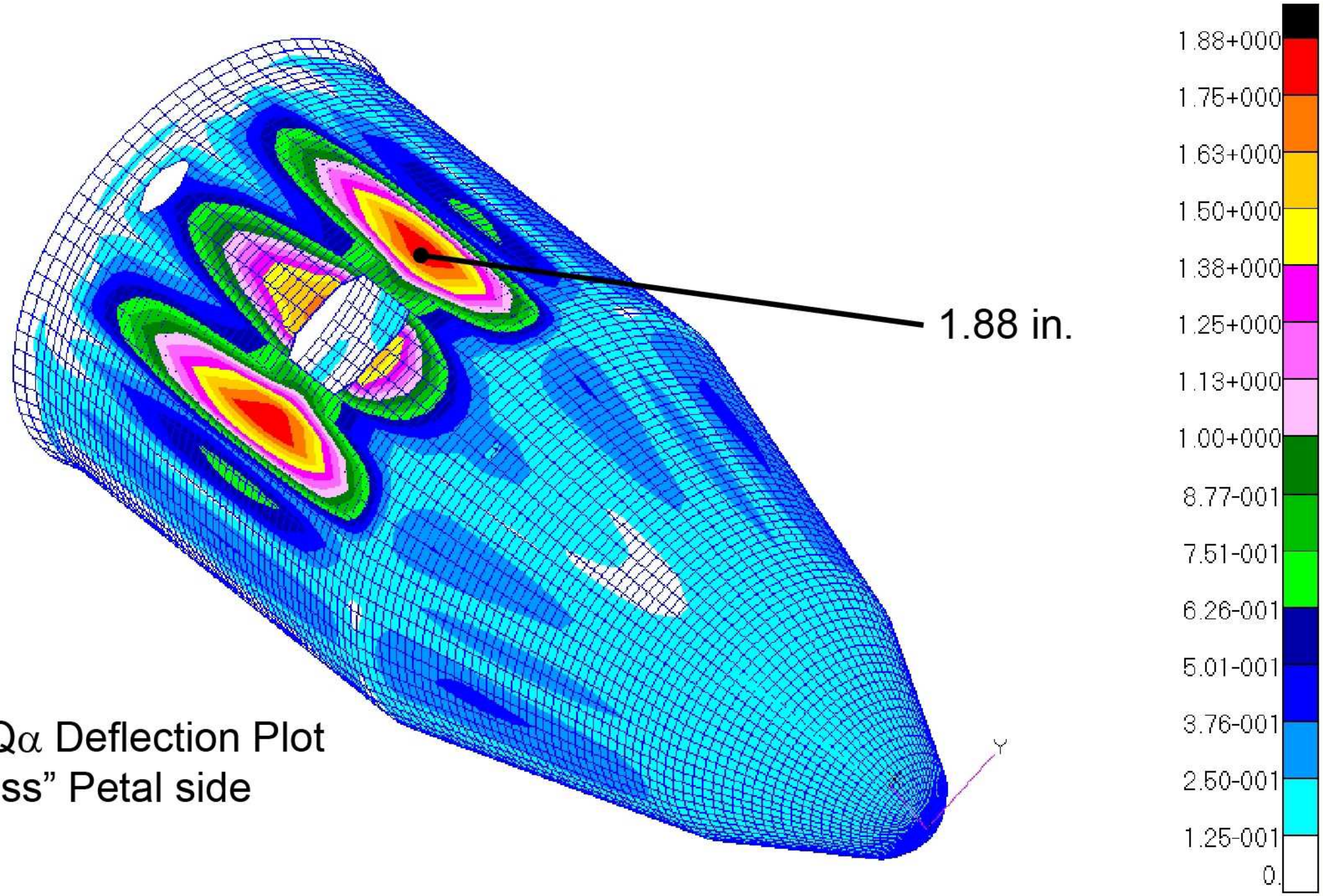
Preliminary Aerothermal Analysis

Mission Maximum Temperature



Preliminary Structural Analysis

Maximum Static Deflection

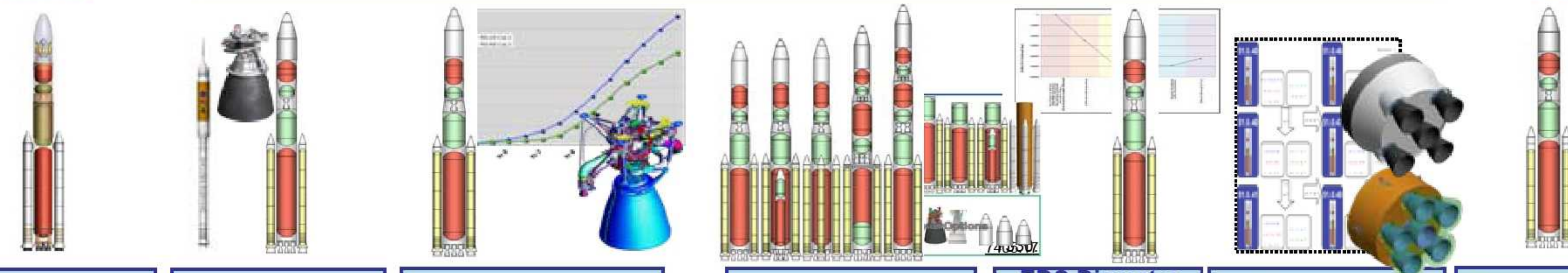


Max Qα Deflection Plot
"Access" Petal side

Ground Rules and Assumptions

- **All trajectories analyzed using POST3D (Program to Optimize Simulated Trajectories - 3 Dimensional)**
- **Flight performance reserve is based on the Ares V LEO mission, and is held constant for all cases**
- **No gravity assists**
- **Interplanetary trip times are based on Hohmann transfers (limited to ~24 years max.)**
- **Payload mass estimates are separated spacecraft mass, and include payload adapter and any mission peculiar hardware (if required)**
- **Ares V vehicle based on configuration 51.00.39, but w/ Upper Stage burnout mass from configuration 51.00.34 (propellant tanks not resized for high C3 missions)**
- **For cases incorporating a kick stage:**
  - Ares I and Ares V employ 2-engine Centaur from Atlas V
  - Additional adapter mass of 6,400 lbm assumed
  - No adjustments to aerodynamic data
- **Propellant mass for:**
  - Ares V LEO missions: held constant at 310,000 lbm
  - Ares I and V C3 missions and Ares I LEO missions: maximum propellant load
- **No Upper Stage propellant off-loading for Ares I and Ares V C3 cases**
- **Transfer orbit to Sun-Earth L2 point is a direct transfer w/ C3 = -0.7 km2/s2**
  - Payload can be increased by using a lunar swingby maneuver
- **All cases targeting a C3 are of longer duration than the J-2X constraint of 500 seconds**

ESAS (2005) to LCCR (2008) Major Events



Original ESAS Capability
- *45.0 mT Lander*
- *20.0 mT CEV*
- *No Loiter in LEO*
- *8.4m OML*
- *5 SSMEs / 2J2S*

CY-06 Budget Trade to Increase
- Ares I / Ares V Commonality
- Ares I : 5 Seg RSRB / J2-X instead of Air-Start SSME
- Ares V: 1 J2-X

Detailed Cost Trade of SSME vs RS-68
- ~$4.25B Life Cycle Cost Savings for
- 5 Engine Core
- Increased Commonality with Ares I Booster
- 30-95 Day LEO Loiter Assessed

IDAC 3 Trade Space
- Lunar Architecture Team 1/2 (LAT) Studies
- Mission Delta V's increased
- Increase Margins From TLI Only to Earth through TLI
- Loiter Penalties for 30 Day Orbit Quantified

EDS Diameter Change from 8.4m to 10m
- Lunar Architecture Team 1/2 (LAT) Studies
- Lunar /Mars Systems Benefits
- Tank Assembly Tooling Commonality

Incorporate Ares I Design Lessons Learned / Parameters
- Core Engine / SRB Trades to Increase Design Margins
- Increase Subsystem Mass Growth Allowance (MGA)

Recommended Option
- *6 Core Engines*
- *5.5 Segment PBAN*

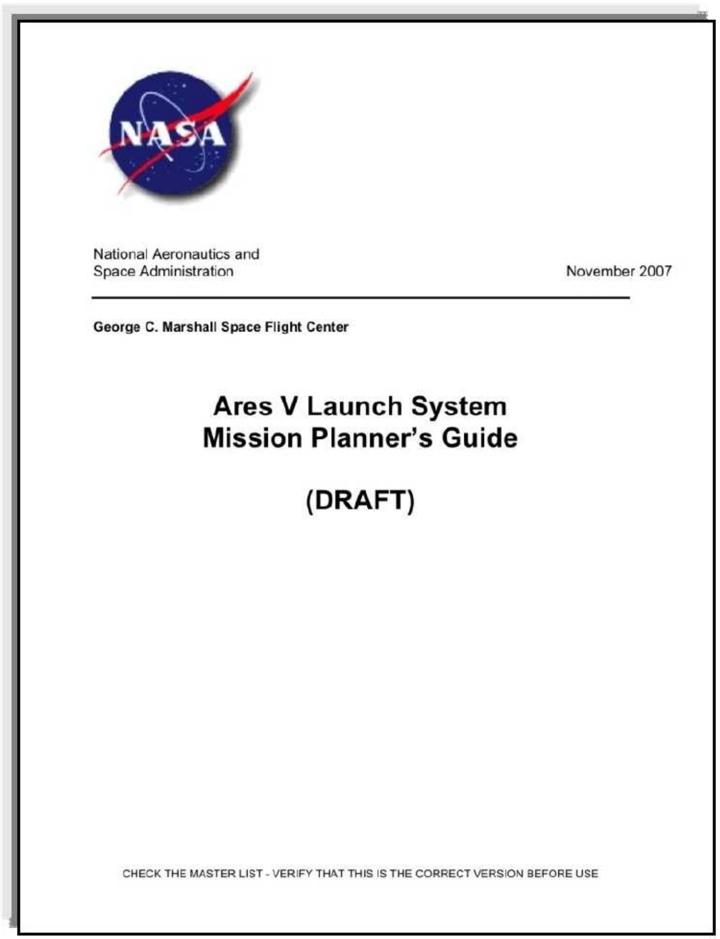
Updated Capability
- 45.0t Lander
- 20.2t CEV
- ~6t Perf. Margin
- 4 Day LEO Loiter
- Ares I Common MGAs
- Booster Decision Summer 2010

220 Concepts Evaluated | 320 Concepts Evaluated | 730 Concepts Evaluated | 460 Concepts Evaluated

2005 | **2006** | **2007** | **2008**

Ares I ATP — Orion ATP — Ares I SRR — Orion SRR — Ares I SDR — Ares V MCR

Ares V Launch System
Mission Planner's Guide


National Aeronautics and
Space Administration

November 2007

George C. Marshall Space Flight Center

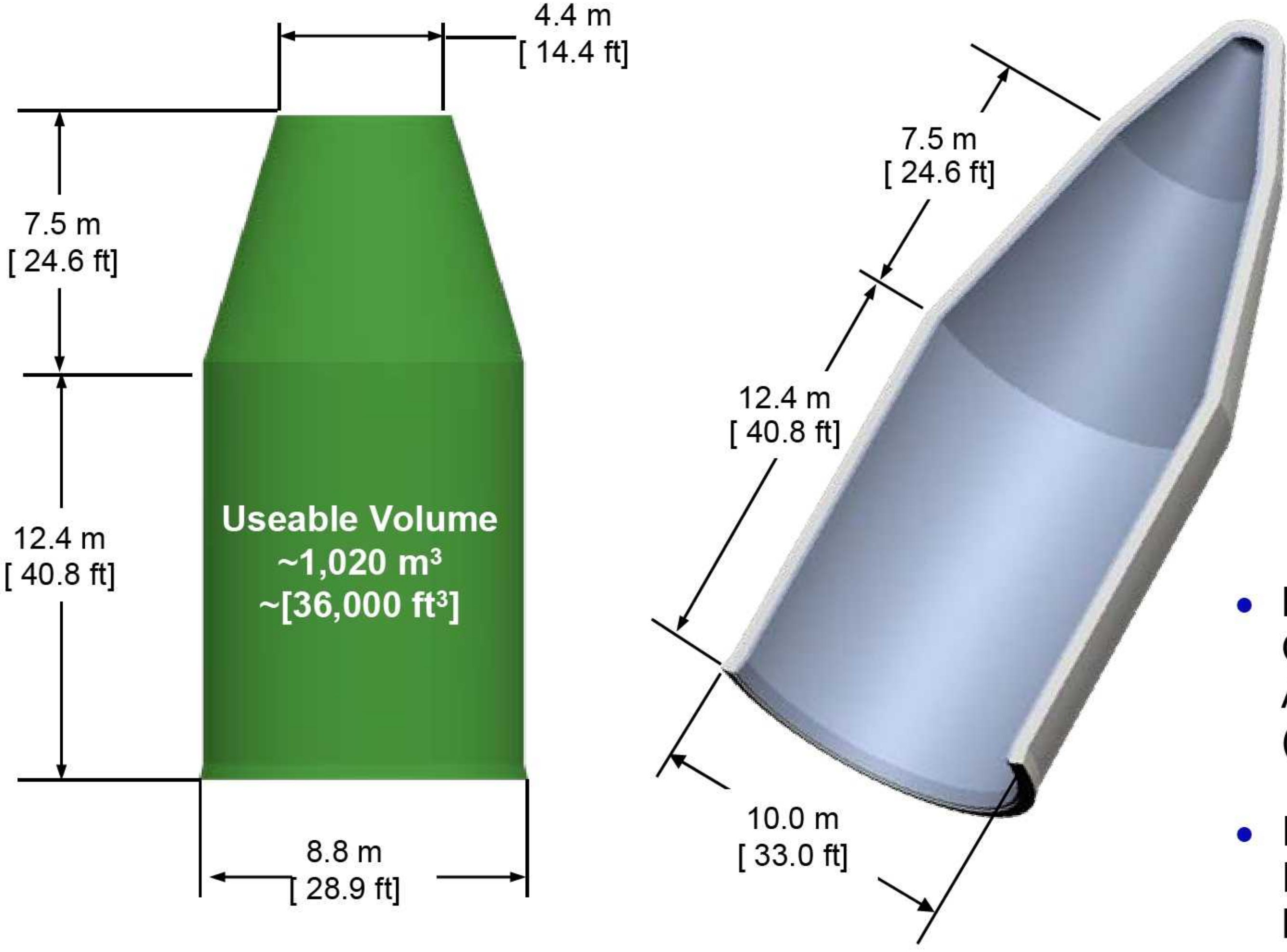
Ares V Launch System
Mission Planner's Guide

(DRAFT)

CHECK THE MASTER LIST - VERIFY THAT THIS IS THE CORRECT VERSION BEFORE USE

- **Mission Planner's Guide In Development**
  - Interface Definitions
    - Fairings, Adapters…
  - Mission Performance
  - Development Timelines
  - Concept of Operations
  - Potential Vehicle Evolution and Enhancements
  - Need Past Astronomy Mission Data
  - Based on 51.00.39 concept

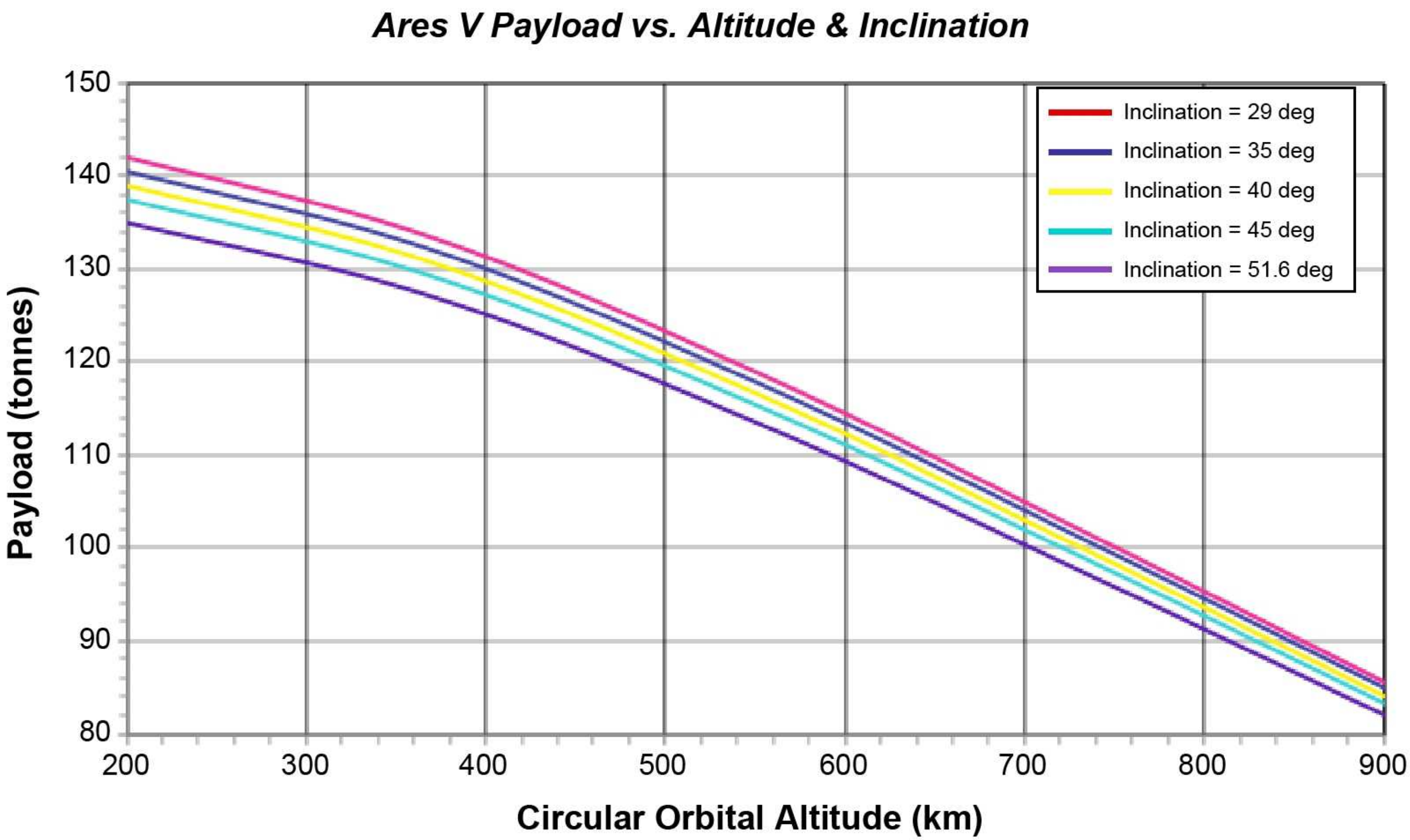
Notional Ares V Shroud for Other Missions



- Maximum Barrel Length Constrained Vehicle Assembly Building (VAB) Height
- Increased Barrel Length Introduces Theoretical Reduction of Payload Capability of 200 kg

Ares V LEO Performance

Extended Shroud



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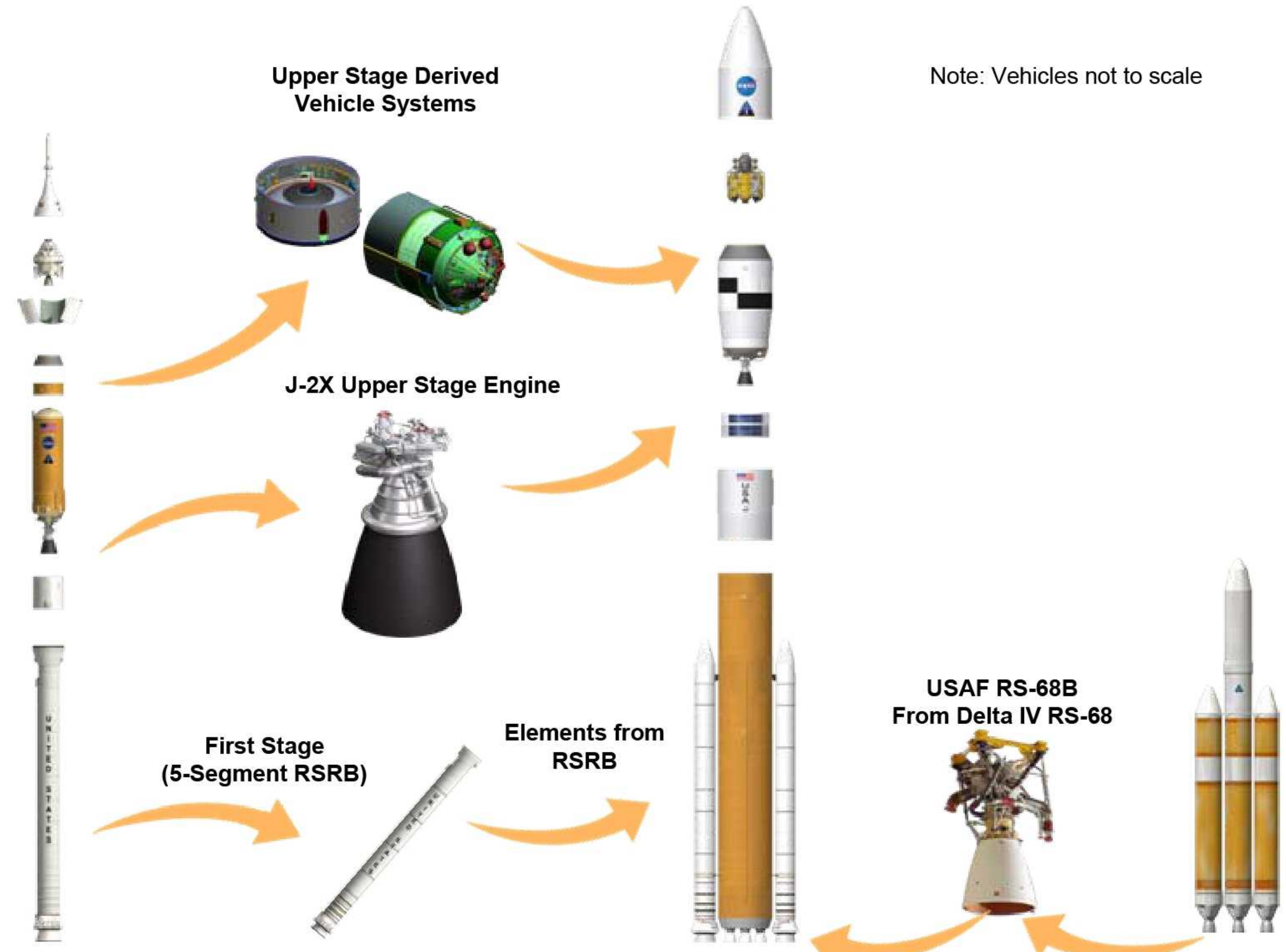
Employing Common Hardware to Reduce Operations Costs



Ares I
25.5 mT (56.2K lbm) to Low Earth Orbit (LEO)

Ares V
71.1 mT (156.7K lbm) to TLI (with Ares I)
63.0 mT (138.5K lbm) to Direct TLI
187.7 mT (413.8K lbm) to LEO

Delta IV