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X-38 Application of Dynamic Inversion Flight Control

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This paper summarizes the application of a nonlinear dynamic inversion (DI) flight control system (FCS) to an autonomous flight test vehicle in NASA's X-38 Project, a predecessor to the International Space Station (ISS) Crew Return Vehicle (CRV). Honeywell's Multi-Application Control-H (MACH) is a parameterized FCS design architecture including both model-based DI rate-compensation and classical P+I command-tracking. MACH was adopted by X-38 in order to shorten the design cycle time for different vehicle shapes and flight envelopes and evolving aerodynamic databases. Specific design issues and analysis results are presented for the application of MACH to the 3rd free flight (FF3) of X-38 Vehicle 132 (V132). This B-52 drop test, occurring on March 30, 2000, represents the first flight test of MACH and one of the first few known applications of DI in the primary FCS of an autonomous flight test vehicle.

INTRODUCTION

X-38 Mission Overview and Requirements

The NASA Johnson Space Center (JSC) X-38 Project is a technology demonstrator and prototype for the CRV, a "lifeboat" for ISS. The CRV will be capable of returning an injured or incapacitated astronaut to earth in a medical emergency, or all seven ISS crew members during an evacuation. The CRV could be modified for other uses, such as a general crew transport vehicle launched atop an Ariane 5 booster. It will become the first new human spacecraft to travel to and from orbit in the past two decades, developed at a fraction of the cost of past manned space vehicles. The reduced development cost and accelerated schedule made the X-38 Project attractive given current NASA budgetary constraints and ISS CRV requirements.

The X-38 Project includes multiple lifting body aircraft based largely upon airframes tested by the U.S. Air Force in the 1960s. X-38 flight testing began with subsonic vehicles air-launched from the starboard wing of a B-52 bomber at NASA's Dryden Flight Research Center (DFRC), and will culminate in a space vehicle which will be deployed from a Shuttle cargo bay for a full deorbit, entry and landing test. Vehicle 131 (V131) was uncontrolled, used to test the aerodynamics and parachute and landing systems. V132 was the first X-38 test vehicle with an autonomous FCS. The first two free flights (FF1 and FF2) of V132 used a classical derivation of the Shuttle entry FCS. This report focuses upon V132 FF3, which employed MACH, a Honeywell implementation of DI.

Technical Issues Raised

X-38 drop tests present many challenges for FCS design. X-38 lifting bodies have inherently coupled, nonlinear dynamics and reduced controllability and stability relative to traditional aircraft. The B-52 bomber used as a drop test carrier aircraft imparts strong, transient aerodynamic moments upon the test vehicle during the first couple of seconds following release. The fact that FF3 was the fastest and highest of the three V132 flight tests only magnified this B-52 transient as a design issue. MACH's computational requirements had to fit in the FCS budget on the V132 flight computer. The nonlinear MACH controller had to be linearized in order to apply linear stability analyses and verify stability margin requirements were being met across the range of large aerodynamic uncertainties. An estimate of sideslip angle was necessary to meet MACH's full-state feedback requirement.

MACH OVERVIEW

MACH is a blended implementation of classical and modern controllers. Classical proportional + integral (P+I) outer loops provide attitude angle command tracking, while DI inner loops provide rate compensation.

Applications

MACH is a decade old, but had not reached the flight test stage before X-38. V132 FF3 represented the first flight test of MACH, and, the authors believe, one of the first few flight tests of a primary autonomous FCS using DI as a central design technique.

Program/Vehicle	Years	Current Maturity
F-18 High Angle-of-attack Research Vehicle (HARV)	1991-1995	HW-in-the-loop sim.
Joint Defense Attack Munition (JDAM)	1992	Engineering sim.
DC-X, high-AOA atmospheric maneuvering	1993	Engineering sim.
F-117, USAF Multivariable Control Design Guidelines	1993-1996	Engineering sim.
YF-22, USAF Multivariable Control Design Guidelines	1993-1996	Engineering sim.
MCT/F-16, USAF Multivariable Control Design Guidelines	1993-1996	Engineering sim.
Lockheed-Margin Joint Strike Fighter (JSF), Marine STOVL	1995-1999	Piloted sim.
X-38 V132 FF3	1996-1999	Flight test
X-38 V131R	1999-2001	Flight test
X-38 V201	1998-2002	Engineering sim.

Table 1: History of MACH Applications

Dynamic Inversion

DI is a multivariable, nonlinear, model-based controller design technique. The idea is not new, proposed for FCS applications as early as 1975 by Meyer and Cicolani (NASA TN D-7940). However, considerable computational requirements made DI FCS applications impractical until the last decade.

DI Basics

1. Compute a model of the vehicle's nonlinear dynamics in real time during flight.
2. Dynamically invert this model to cancel expected nonlinear dynamics.
3. Insert desired vehicle response dynamics, which are typically linear and 1st order, e.g., $k / (s + k)$.

Disadvantages of DI and MACH

1. It can be *computationally intensive* for a flight computer to compute and invert a nonlinear vehicle model in real time. This is largely the reason MACH has not flown before X-38. Several streamlining modifications were made to MACH in order for its computational requirements to fit within the FCS timing allotment on the V132 flight computer.
2. The on-board model of vehicle dynamics requires *full-state feedback*. So, vehicle states must be measured by a full complement of sensors, and those states that are not measured must be estimated, e.g., sideslip angle.
3. DI can be *complex to implement* in software. The general version of MACH has added complexity due to its "multi-application" feature, which in part means it is automatically scalable for different numbers and types of control effectors and states. The computational speed enhancements mentioned above included stripping out some of this general functionality and customizing MACH specifically for V132.

Advantages of DI and MACH

1. Preliminary FCS designs for various, similar vehicles may be realized simply by swapping out the on-board model of vehicle dynamics, keeping the overall architecture largely unchanged. MACH's generality of application was well suited to the X-38 Project, which required FCS designs for multiple, similar vehicles in a tight schedule.
2. The DI portion of MACH reduces or even eliminates the need for point design gain scheduling. Traditional FCS design techniques involve tailoring a series of point designs to the vehicle-specific aerodynamics at pre-selected flight conditions along a trajectory or within a flight envelope, then linearly scheduling gains between these points. DI may be described simply as automatic, real-time, nonlinear gain scheduling, based upon the same vehicle aerodynamics database used to tune traditional point designs. So, for the X-38 test vehicles, there is no gain scheduling in the traditional sense, though MACH's controlled variable structure and outer loop gains must be tuned for each subsonic vehicle, and for each control mode of the entry vehicle, e.g., blended jets and aerodynamic effectors, no rudder during high heating, etc.
3. DI compensates for any nonlinear, coupled dynamics included in the on-board computational model of the vehicle. X-38 lifting bodies have some nonlinear aerodynamics, e.g., yawing moment as a function of sideslip angle, and have much more highly coupled longitudinal and lateral-directional dynamics than traditional, winged aircraft.

Architecture

MACH's overall structure (Figure 1) includes concepts ranging from conventional to very modern.

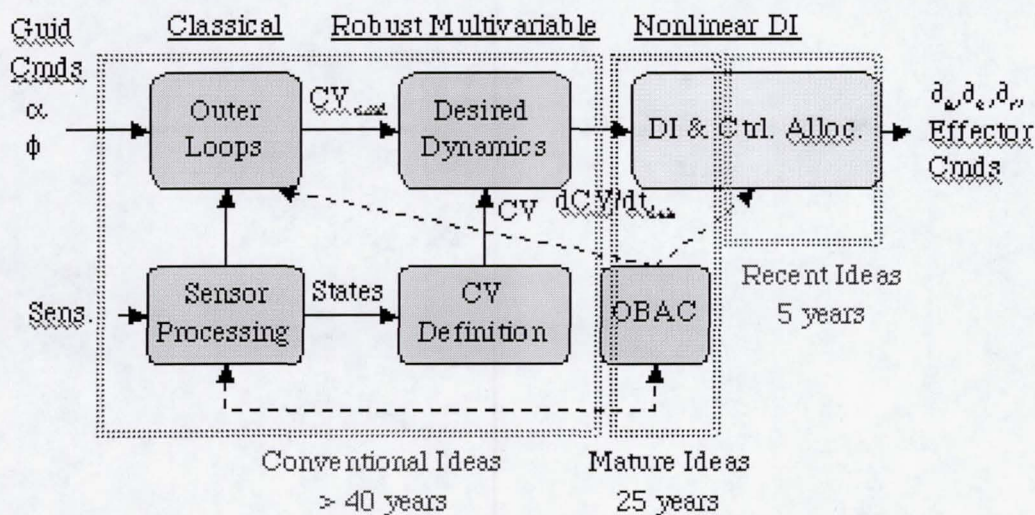


Figure 1: MACH Architecture

There are five main ingredients in MACH:

1. **Controlled Variables (CV)** are selected for performance (command tracking and disturbance rejection) and robustness. They must offer stable zero dynamics, as inverted zeros become poles in DI. CV definitions may blend rotational and translational control, and provide coordinated flight in a steady wind. V132 MACH uses three CVs, LCV, MCV and NCV for roll, pitch and yaw axes, respectively.
2. **Outer loops** convert guidance commands into CV commands (Figure 2), and include trim integrators to protect for vehicle modeling errors. In V132 MACH, LCV_{cmd} depends upon the bank angle command, MCV_{cmd} depends upon the commanded angle of attack (α), and NCV_{cmd} is zero, indicating a desire for zero sideslip angle (β).

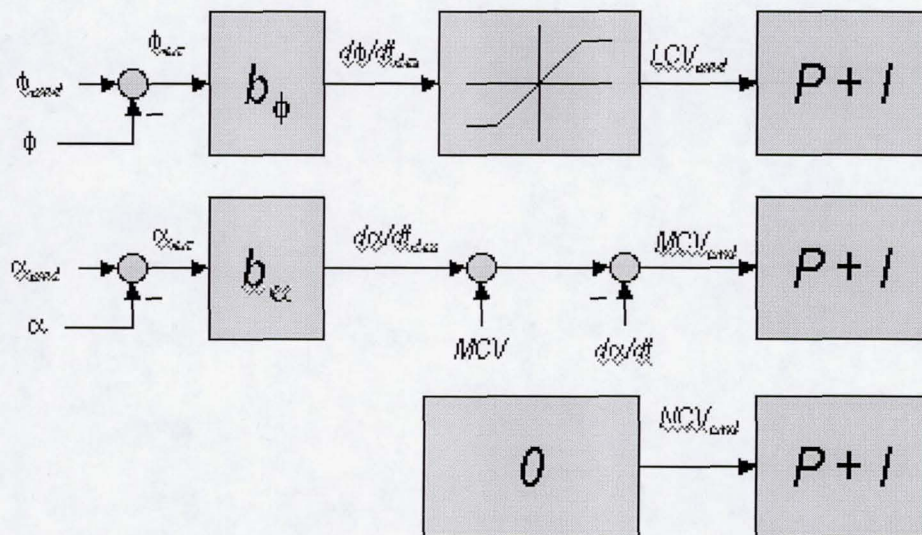


Figure 2: MACH Outer Loops

3. **Desired dynamics** are defined by P+I elements, and may be described by a 1st order transfer function, assuming perfect DI reduces all inner loop dynamics to a simple integrator, $1/s$.

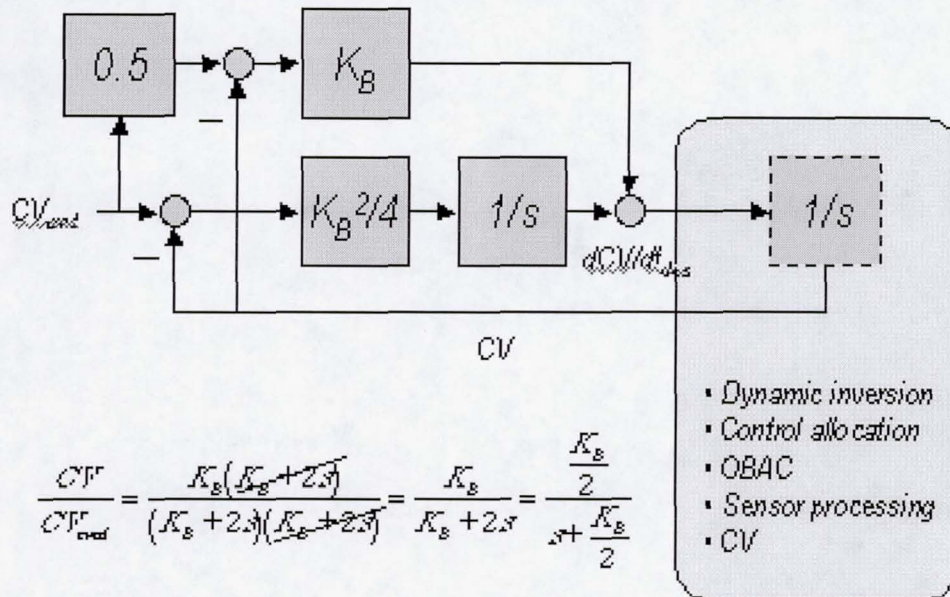


Figure 3: MACH Desired Dynamics

4. **Control allocation** optimally divides control authority between effectors when shared surfaces are on rate or position limits, based upon preset axis priority weighting, e.g., $NCV/LCV = 5$ implies NCV control is five times as important as LCV.
5. The **On-Board Aircraft Model (OBAC)** computes the nonlinear vehicle dynamics on the flight computer. In the interest of real-time computational efficiency, a least-squares (LSQ) method was used to approximate the full V132 aerodynamics database. The result was a compact, fast LSQ aerodynamics model containing only 175 values at a constant mach number of 0.6 and trim elevon deflection of 30° , nonlinear with respect to α but linear with respect to angular rates (p, q, r) and aerosurface deflections, aileron, elevon and rudder ($\delta_a, \delta_e, \delta_r$).

VEHICLE DESCRIPTION

X-38 Vehicles

X-38 airframes evolved from the X-24 wingless lifting body test vehicles of the late 1960s. The first two X-38 subsonic test vehicles, V131 and V132, closely resembled the X-24A shape, which enabled existing X-24A aerodynamics data to reduce the need for early X-38 wind tunnel testing. The planned space flight test vehicle, V201, and its sub-scale atmospheric equivalent, V131R, expand the V131/132 shape to provide the additional space necessary for seven ISS crew members, two drogue chutes and a larger main parafoil. X-38 vehicles have four aerodynamic control surfaces, two rudders and two body flaps. Rudder deflection (δ_r) provides roll and yaw control. Though not used for V132, the rudders also may be moved in opposite directions for “speed brake” deflection, which can be used to augment pitch trim. Moved in the same direction, the two body flaps provide elevon deflection (δ_e) for pitch control, and when moved in opposite directions they provide aileron deflection (δ_a) for limited roll control. The subsonic test vehicles are single string systems, while the V201 space test vehicle will have multiple redundancy.

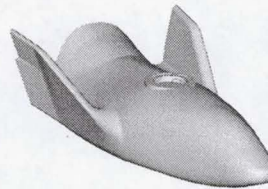


Figure 4: V131/132 Figure 5: V131R/201

Vehicle	V131	V132	V131R	V201
Primary FCS	None	FF1&2: Classical, FF3: MACH	MACH	MACH
Mass (lbs)	14000		18000	25000
Main Structure	Fiberglass		Fiberglass	Aluminum
Length (in)	283		283	340
Width (in)	120		120	144
Xcg (% Ref. Length)	57.1		59.2	59.0
Reference Length (ft)	23.0		23.0	27.6

Table 2: X-38 Vehicle Properties

V132 GN&C Subsystems

The V132 MACH FCS ran at 25 Hz on a Motorola 68040 processor running at 25MHz. The MACH flight software was a subset of the V132 GN&C executive software that interfaced with the avionics systems. V132 “guidance” was simply a pre-programmed, time-based, open-loop angle-of-attack command profile and a constant bank angle command of zero overlaid with time-based Programmed Test Input (PTI) commands used for post-flight aerodynamic extraction. A Honeywell Embedded GPS/INS (EGI) box provided attitude, rate, and acceleration data. A Flush Air Data System (FADS) located in the vehicle nose cap used differential port pressure measurements to provide relative velocity, α , β , dynamic pressure, and mach number. The four aerodynamic control surfaces were each moved by a single electro-mechanical actuator (EMA).

V132 Vehicle Safety

Although V132 systems were “single string,” i.e., no redundancy, many GN&C systems could “safe” the vehicle if a problem developed. V132 software could initiate early flight termination and drogue deployment if attitude rates or dynamic pressure exceeded predetermined limits. A FADS failure was handled by switching to a backup Navigation Air Data (NAVDAD) algorithm to provide EGI-derived estimates of vehicle relative velocity, α , β , dynamic pressure, and mach number. Communication faults and computational errors were monitored and, if persistent, could result in early flight termination. In the event of a total failure of the flight computer, drogue chute or parafoil, a backup chute could be deployed from the ground. Finally, the vehicle had a completely independent flight termination system triggered by a range safety officer if the vehicle strayed outside of its range boundary.

V132 Subsonic Aerodynamics and Uncertainties

Lifting bodies provide significant challenges to FCS design. Short moment arms result in limited control authority and sluggish vehicle response, especially in roll. In fact, V132 rudders were three times more effective in roll than the ailerons, which exemplifies the coupled nature of V132 control. The blunt aft end induces asymmetric base flow which imparts yaw and roll moments upon the vehicle, as observed during V132 flight tests. Fortunately, these asymmetries were within the uncertainties of the V132 aerodynamic database used for FCS design. Base flow effects also degraded aileron effectiveness.

Vector diagrams showing the contributions of effectors and attitude angles upon moment coefficients were used to assess stability and controllability. Figure 6 contains rudder (left), aileron and β (right) vectors plotted on C_l vs. C_n axes. Uncertainty boxes drawn around each vector were used to determine worst case uncertainty combinations for FCS testing. Wide separation of vectors is desirable for controllability. An ideal aircraft might have a nearly horizontal rudder vector and a longer, nearly vertical aileron vector, i.e., pure roll and yaw axis controllers. Uncertainties or flight conditions which align the effector vectors result in loss of effector independence—redundant effectors—and degraded controllability. Directional stability is reduced as the β vector moves left. β vector separation is required for bank angle controllability.

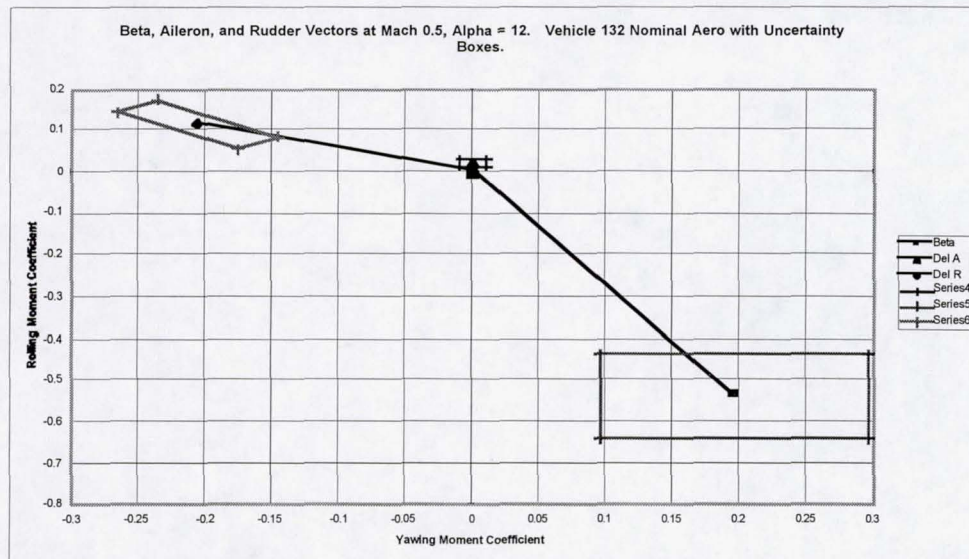


Figure 6: V132 Lateral-Directional Vector Diagram, $M = 0.5$, $\alpha = 12^\circ$

Figure 6 shows the V132 vector diagram at $M = 0.5$ and $\alpha = 12^\circ$. The relatively small aileron vector indicates the low effectiveness of the aileron compared to the rudder. The nominal aileron and rudder vectors are reasonably separated, but there are clearly points on the uncertainty boxes which align the two vectors, so the FCS had to be robust to this alignment case. The aileron and rudder vectors became more closely aligned near $M = 1$. However, the highest, fastest V132 drop test, FF3, began at $M = 0.6$.

Lateral control departure parameters (LCDP) [1] for each effector are measures of bank control and numeric indicators of effector/ β vector alignment. The following equations define aileron and rudder LCDP in terms of stability derivatives, and provide inequality criteria for proverse bank angle control.

$$\text{Aileron: } LCDP_{\delta a} = C_{n\beta} C_{l_{\delta a}} - C_{l\beta} C_{n_{\delta a}} > 0$$

$$\text{Rudder: } LCDP_{\delta r} = C_{n\beta} C_{l_{\delta r}} - C_{l\beta} C_{n_{\delta r}} < 0$$

LCDP = 0 indicates effector/ β vector alignment and neutral bank control, i.e., any control deflection is offset by β , resulting in zero roll. An adverse LCDP, i.e., reversal of either of the above inequalities, indicates reversal of the corresponding effector's bank control authority. For example, while positive aileron deflection normally results in positive roll, a negative aileron LCDP means that positive aileron combines with β to create negative roll—this is called "aileron reversal."

V132 MACH DESIGN

This section describes the control system design modifications developed to address the unique issues associated with the 132 vehicle and flight test. Within the structure of the existing MACH architecture, bandwidth and CV selections were made to satisfy mission, flying qualities and stability margin requirements as stated in the previous section of this paper. Generally, this involved an iterative approach of gain selections tested in linear and nonlinear simulation. In addition, new algorithm changes evolved to handle uncertain aerodynamics, the lack of a sideslip measurement and the large but short disturbance from B52 release. More in-depth discussion of these may be found in references 1-3.

CV Selection

With good separation in elevon, aileron and rudder effectiveness for the anticipated flight near $\alpha = 12^\circ$, three CVs—LCV, MCV, and NCV—were chosen to independently control the three rotational axes—roll, pitch and yaw, respectively. The CVs definitions evolved from previous experience with classical flight control systems and other MACH applications (Table 1), using EGI body attitude, inertial rates, translational accelerations, and FADS measurements of α , velocity and dynamic pressure:

$$LCV = p_s$$

$$MCV = q + \frac{1}{400} \left(\frac{2.5}{s + 2.5} \right) n_z + \frac{g \cos \gamma \cos \mu}{V \cos \beta} - p_s \tan \beta - 0.003 \bar{q} - \left(\frac{g}{400} + \frac{g}{V} \right) \cos \gamma$$

$$NCV = r_s - \frac{\bar{q}}{228} \beta_{equiv} - \frac{g}{V} \cos \gamma \sin \mu$$

These were chosen to stabilize zero dynamics, complement α and ϕ command tracking, and maintain coordinated flight in a steady wind. In NCV, a complementary filter approach (Figure 7) blended low-frequency lateral acceleration (n_y) with high-frequency estimated β (β_{est}) to form an equivalent β (β_{equiv}) which would maintain zero sideslip in steady-state flight. The filter had a bandwidth of 0.5 rad/sec, and was inverted as part of the OBAC.

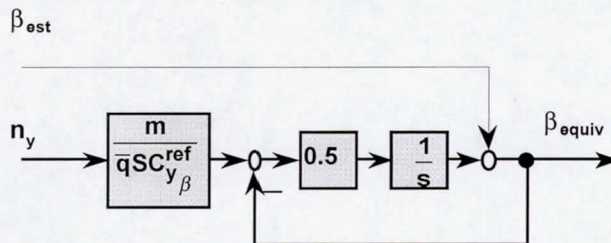


Figure 7: NCV Filtered Sideslip Angle

Loop Design

Inner loop P+I compensation bandwidths were set to 1.0 rad/sec for LCV and NCV and 1.7 rad/sec for MCV by setting the K_B design parameter to 2 and 3.4, respectively (see Figure 3). Anti-windup logic was employed to prevent integrator build-up during periods of surface saturation.

The outer loops turn α and ϕ commands into MCV and LCV commands, respectively (Figure 2). Outer loop gains, $b_\phi = 0.4$ and $b_\alpha = 1.0$, were based upon performance requirements and inner loop bandwidths. Note that angle-of-attack rate is computed in the OBAC and not sensed.

Beta Estimation

In general, MACH requires full-state feedback, and therefore requires a sideslip angle sensor for OBAC computations. Because FADS-computed sideslip angle was not available for feedback use on V132, four β estimation options were investigated: (1) special cases of Kalman filters, (2) inertial sideslip, (3) sideslip derived from lateral acceleration, and (4) zero sideslip. Option 1 was eliminated early because it was oversimplified and never tuned properly. Option 2 was unacceptable in steady winds where it is destabilizing at low frequencies. Option 3 led to poor stability margins. Option 4 ultimately was chosen.

Further rationale for option 4 came from a stability margin analysis for a typical flight condition and dynamic pressure. The analysis considered a low-passed measurement of sideslip angle as the control law sensor. The low pass filter break frequency was varied between 0 and 100 rad/sec. The best phase and gain margins were achieved for low break frequencies (Figure 8). In fact, approximating sideslip as zero gave the best robust stability. Also, the minimum closed-loop damping ratio is only slightly degraded with this approximation. Thus, the simplest option was selected, and the estimate of sideslip was set to zero.

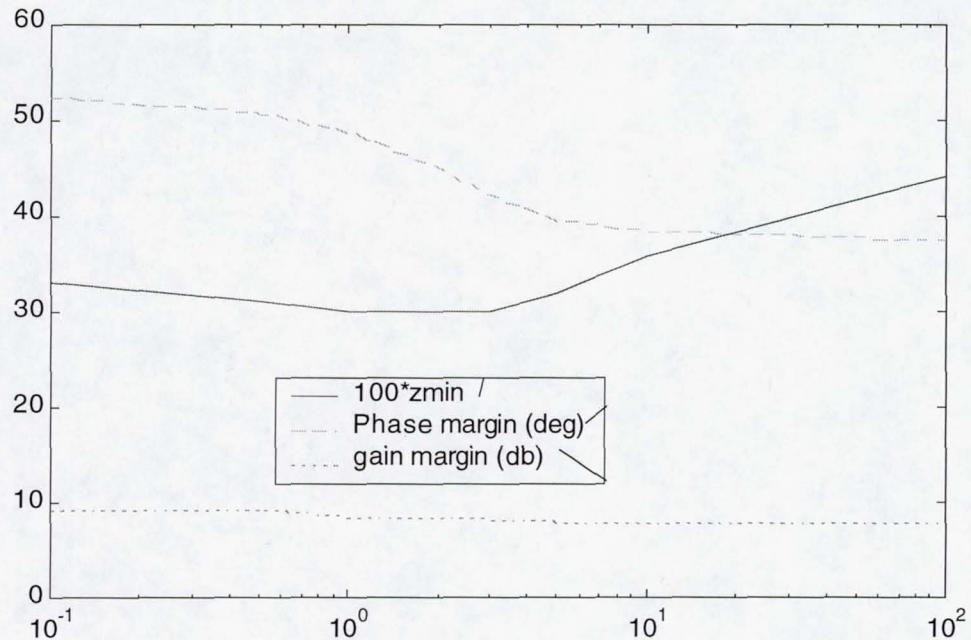


Figure 8: Nominal and Robust Stability Analysis of Low-Pass Filtered Sideslip

B-52 Transient

When V131 and V132 separated from the starboard wing of the B-52, they experienced a strong downward pitch and positive roll and yaw moments due to the sway brace flow interference and down-wash and out-wash effects from the B-52 wing and fuselage. These dissipated quickly as the distance from the B-52 increased; however, for V132, the resulting high angular rate transient had to be actively controlled. Though V132 control effectors could not overcome these forces, they were used at saturation limits to mitigate the disturbance as much as possible.

Due to the highly nonlinear nature of this B-52 transient problem, 6-DOF simulation was used heavily to converge upon a viable solution. Nearly 80% of the control design effort was incurred solving this problem, primarily through trial and error. The solution that evolved consisted of two changes to the control algorithm and two associated with its initiation. These 4 changes are now presented.

Control Allocation

The control allocation logic in MACH is based upon specified effector limits and preferences. Typically, when shared surfaces such as the body flaps are used for both elevon and aileron deflections, preference would be given to the elevon to satisfy the MCV (or angle-of-attack) command. Any remaining authority would then be available to the aileron to help satisfy the LCV (or bank angle) command. This scheme proved to be problematic when controlling the B-52 disturbance: All available flap motion was allocated to the elevon, preventing adequate roll authority. As a solution for V132, 50% of the flap rate and 4° of flap position was reserved for use by the aileron when necessary. When not required by aileron, the reserve was made available back to the elevon.

Figure 9 illustrates this “aileron reserve.” The diamond bounds achievable elevon (vertical) and aileron (horizontal) deflections given that elevon is limited by the flap range of 0 to 40° and aileron, while physically able to reach 20° in magnitude, is limited to 15° in software due to available aerodynamic data. Aileron and elevon commands inside these limits will be achieved. For commands outside these limits, the pitch axis has priority except at the top and bottom corners of the diamond. Pitch priority is indicated by horizontal lines mapping commanded points outside the diamond to achievable points on or within the diamond. The 4° of aileron reserve is indicated in the top and bottom corners by the vertical lines mapping points outside the diamond to achievable points on the diamond. Thus, Figure 9 shows how desired elevon and aileron commands lying outside position limits are mapped to achievable commands.

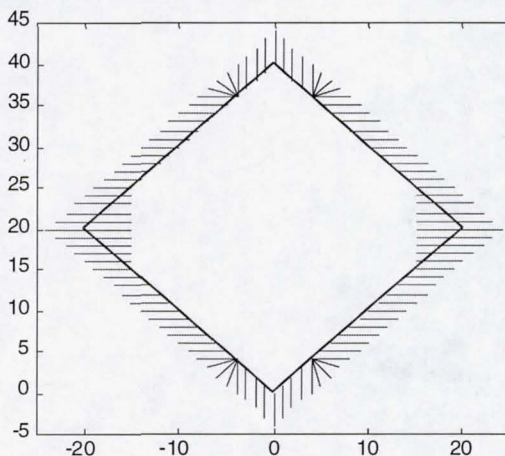


Figure 9: Achievable Elevon-Aileron Diamond

When V132 control surfaces hit rate or position limits and desired LCV and NCV rates could not be satisfied, MACH DI and a control allocation algorithm known as SMN (ref 7) used priority weighting to emphasize NCV errors 5 times more than LCV errors in establishing surface commands that approximately satisfied CV commands, i.e., “zero sideslip” took priority over “wings level” during the B-52 transient.

Delayed Integrator Activation

The desired dynamics used classical P+I compensation for frequency shaping and bandwidth selection. The NCV integrator charged up during the B-52 transient despite the anti-windup logic and in some instances caused an undesirable negative roll overshoot, i.e., an “over-reaction” to the initial positive roll transient. This was remedied by removing integral action in the NCV channel for the first 3 seconds of flight. When the NCV integrator was activated at 3 seconds, the transition was smooth with no transient.

Control Law Activation Time

The time at which the FCS was activated after release was a design parameter used to manage the B-52 transient. Though the vehicle airframe was stable, it was determined that early activation of the control law was best. DFRC analysis confirmed there was no recontact issue, so MACH was activated at the earliest possible time of 0.32 seconds after release.

Initial Elevon Position

The transient negative pitch rate caused the elevon to initially move upward, pulling the nose back up. The downward pitching moment quickly disappeared, and the elevon, riding the rate limit, could react quickly enough to avoid an overshoot in angle-of-attack, often above the 20° “comfort zone” limit where rudder stall was feared. Setting the initial position at a lower (relative to previous drop tests) deflection of 25° reduced this α overshoot. Moving the elevon near the middle of its range also increased available aileron deflection. Special MACH initialization software was developed to load the MCV integral state and produce an equal elevon command, thus avoiding a startup command error transient.

Robustness to Aerodynamic Uncertainties

MACH architecture enables frequency loop-shaping in the desired dynamics block to provide robust stability in the presence of high-frequency plant uncertainties (ref. 4). However, bandwidth selection alone provided insufficient robustness, i.e. stability margins, over the large range of V132 aerodynamic uncertainties. A new tack to improve robustness involved the selection of the aerodynamic model used in the OBAC. Instead of a nominal aerodynamic model, a particular aerodynamic uncertainty set was selected to describe the dynamics in the OBAC. The result was increased robust stability with a small expense to nominal performance. The selection procedure is described below for the linear multivariable case and demonstrated with a first order, SISO example.

Multivariable Motivation and Solution

Figure 10 is a block diagram of the innermost dynamic inversion loops. The problem is to achieve robust stability with respect to particular models which are spread around the set of uncertainty models. The eigenvalues of the closed loop in Figure 10 will be used for model selection. The subscript i indexes the particular state-space model of the aircraft and j indexes the model used in the control law OBAC.

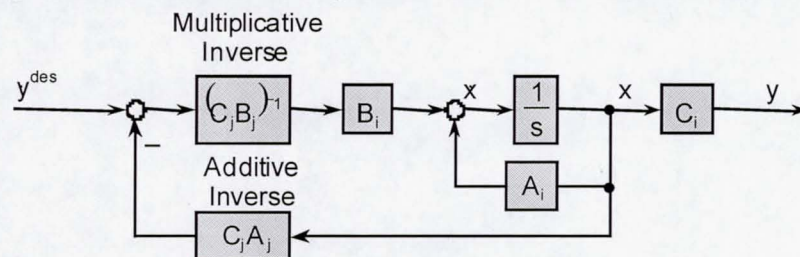


Figure 10: Block Diagram of Closed Dynamic Inversion Loops

In the ideal case where there is no model mismatch ($A_i = A_j$, $B_i = B_j$, $C_i = C_j$), an analytical derivation of the closed loop transfer function from \dot{y}^{des} to y shows pole/zero cancellations which result in a pure integrator. Now when there is model mismatch, the poles and zeros do not cancel, and some of the remaining poles may be unstable. An instability may be fast enough that it will not be robustly stabilized by the other, outer control loops supplied by MACH.

The solution presented here is to pick the control law model that best stabilizes—i.e., has the most negative real part of the eigenvalues—all aircraft models—measured by the most unstable eigenvalue. In general, this involves solving for the eigenvalues of i^*j closed loop systems. For any (i,j) pair, the most unstable eigenvalue is given as:

$$\lambda_{ij} = \max \text{Re} \{ \text{eig} [A_i - B_i (C_j B_j)^{-1} C_j A_j] \}$$

Then the rule is to choose the control law model (index j) corresponding to

$$\Lambda = \min_j \max_i \lambda_{ij}$$

Note that this solution assumes perfect sensors and actuators, and full state feedback.

SISO Example

Consider a first order dynamic model, $\dot{x} = ax + bu$, where $-3 \leq a \leq -1$ and $1 \leq b \leq 3$. The parameter space and five specific models of interest are shown in Figure 11. This is a typical scenario in which the corners of the parameter space are specified and the nominal is assumed to be at the middle.

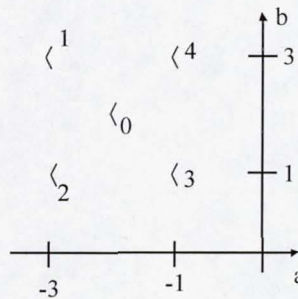


Figure 11: Parameter Space for First Order SISO Example

Computing closed-loop eigenvalues for all combinations of plant and control yields the table in Figure 12. Note that the figure also shows the steps for model selection. First, for each control law, close the loop with all possible aircraft models and note the most unstable eigenvalue. Second, note the most stable of the result of step one. Finally, pick the control law model associated with the result of step two.

		Possible aircraft models					max λ_{kg} across row
		0 (nominal)	1	2	3	4	
Possible control law models	0 (nominal)	0	0	-2	0	2	2
	1	0	0	-2	0	2	2
	2	4	6	0	2	8	8
	3	0	0	-2	0	2	2
	4	-1.3	-2	-2.7	-0.7	0	0

min down the column max across row

Choose model 4 as the control law design model

Figure 12: Model Selection for First Order SISO Example

For this simple example, the best option is to choose point 4 which is something other than the nominal for the control law model.

X-38 V132

The driving requirements for V132 were in the lateral-directional axes. There are nine aerodynamic derivatives of interest in the parameter space: rolling and yawing moment and side force coefficient derivatives with respect to sideslip angle, aileron and rudder. 167 sets of model uncertainty representing a variety of sizes and combinations of parameter uncertainty were considered. As in the simple SISO example above, the nominal aerodynamic model was not chosen, as it did not provide the most robust stability across the dispersion space. The particular uncertainty combination chosen was used to generate the OBAC.

CERTIFICATION PROCESS

V132 MACH FCS certification was the final pre-flight verification that the software met stability and performance requirements. Linear analysis verified gain and phase stability margins, and nonlinear, 6-DOF simulation verified performance and stable oscillations.

Nonlinear Certification

A high-fidelity, 6-DOF simulation of X-38 vehicle dynamics with closed loop GN&C systems was used to look for V132 FCS instabilities and performance violations in a variety of dispersed stress cases. The X-38 simulation was derived from the Shuttle version of the Spacecraft Engineering Simulator (SES), a mature engineering tool long used for system development and post flight trajectory reconstruction. The SES includes high-fidelity models of the environment (aerodynamics, gravity, winds, and atmosphere), flight hardware (actuators, sensors, control surfaces) and flight software (GN&C executive, FCS, etc.). In addition to all-software simulations, the SES was hooked up to V132 for real-time, closed-loop testing of hardware and software. For nonlinear certification, the SES ran in batch mode, varying deterministic "worst-case" combinations of aerodynamic, sensor, and initial condition parameters.

The V132 MACH nonlinear certification included a series of 18 test cases and 2000 simulated drops the SES. These SES runs verified that V132 would fly well in the given flight envelope, and examined system behavior beyond the expected flight envelope (Table 3).

Parameter	Variation
Alpha (α)	8°, 12° (nominal), 16°
Aerodynamics	34 worst case uncertainty sets
B-52 interference effects	3- σ B-52 aerodynamic interference
Gusts, turbulence	20 fps gusts, moderate turbulence
Mass properties	Variations of X_{cg} , Y_{cg} , Z_{cg} , I_{xx} , I_{yy} , I_{zz}
Winds	99% Edwards Port, Tail, Side, Shear
System failures	1 surface failure, FADS failure.
Initial drop state	Dispersions on initial position, velocity, and heading.
PTI	2 sets of aileron, rudder doublets.

Table 3: Parameter Variations for Nonlinear Certification

For the planned flight trajectory at $\alpha = 12^\circ$, the system met performance requirements and was robustly stable. However, for many lower $\alpha = 8^\circ$ trajectory cases, the system exhibited unstable behavior, often losing roll control while maintaining pitch control. These "stability rolls" were due largely to reversal in V132 aileron control authority at $\alpha < 10^\circ$. Steps were taken to avoid such a low α in the event of a FADS failure, and software was added to start the drogue chute deployment sequence early if necessary.

Linear Certification

Though DI does not guarantee robust stability and performance, the MACH algorithm can be linearized to produce traditional stability margins and linear time responses at steady-state operating points [6]. These were relied upon heavily to evaluate design changes and measure closed-loop robustness. In fact, certification for flight mandated the closed-loop system meet SISO stability margin requirements for selected "worst-case" aerodynamic uncertainty sets. In addition to stability margins, a more thorough multi-input/multi-output (MIMO) approach using structured singular value (μ) techniques was utilized to determine stability and performance robustness over all possible combinations of aerodynamic perturbations. This section summarizes the results of these analyses.

Stability analyses were performed at dispersed flight conditions generated with 33 selected aerodynamic uncertainty sets [ΔC_{l_b} , $\Delta C_{l_{dr}}$, $\Delta C_{l_{da}}$, ΔC_{n_b} , $\Delta C_{n_{dr}}$, $\Delta C_{n_{da}}$, ΔC_{y_b} , $\Delta C_{y_{dr}}$, $\Delta C_{y_{da}}$, ΔC_{m_o}] representing 5- σ statistics along the nominal mission profile. Trajectory “snap shots” were taken at 10, 25 and 40 seconds after B-52 launch where steady-state conditions from the 6-DOF simulation were used to generate linear models. Controller models were built up from linearized OBAC models, created from actual flight code at the sensed flight conditions. Linear plant models included aerodynamic dispersions, sensor and actuator dynamics as well as avionics transport lags. Data characterizing the V132 hardware was obtained through vendor specifications and vehicle end-to-end tests [ref Jay’s web site, even though no one can see it?]. Table 4 summarizes the V132 linear model characteristics. Single loops were broken in 3 actuator (aileron, elevon, rudder) and 8 sensor (p, q, r, V, α , ϕ , θ , n_y) paths to compute open-loop gain and phase margins.

V132 Hardware Element	Linear Model Definition
Accelerometer	1 st order: $\omega_n = 44$ rad/sec
Rate Gyro	1 st order: $\omega_n = 66$ rad/sec
Actuator	Two 2 nd -order: $\omega_n = 50$ rad/sec, $\zeta = 0.7$ $\omega_n = 26$ rad/sec, $\zeta = 0.7$
Transport Lag	40 ms

Table 4: V132 Hardware Models

Results of the SISO linear analyses for the nominal 12 degree angle-of-attack profile are summarized in Figure 13 [ref Jay’s presentation]. Even with 5- σ dispersions, the MACH controller was shown to meet 3- σ stability margin requirements except in one dispersed case in yaw-rate feedback.

A MIMO method for applying all 9 aerodynamic uncertainties simultaneously using μ -analysis was developed to determine worst-case performance and gain and phase margins of the closed-loop system [9]. With this technique, fixed-size uncertainties were used to calculate the worst-case H_∞ gain of the performance transfer matrix. This same worst-case analysis method was used to calculate a conservative measure of the minimum combined gain and phase margins of each input/output. In doing this, the worst-case perturbation set for each channel was automatically determined. The results of these analyses using similar 5- σ dispersions are shown in Tables 5 and 6.

[Tables 5 & 6: show Balas results from AIAA paper, Tables 1 and 2]

X38 FF3 MACH v1.4 Linear Stability Margin Survey (all loops, t=10,25&40s, nominal Alpha Trajectory, 5-sigma aero)

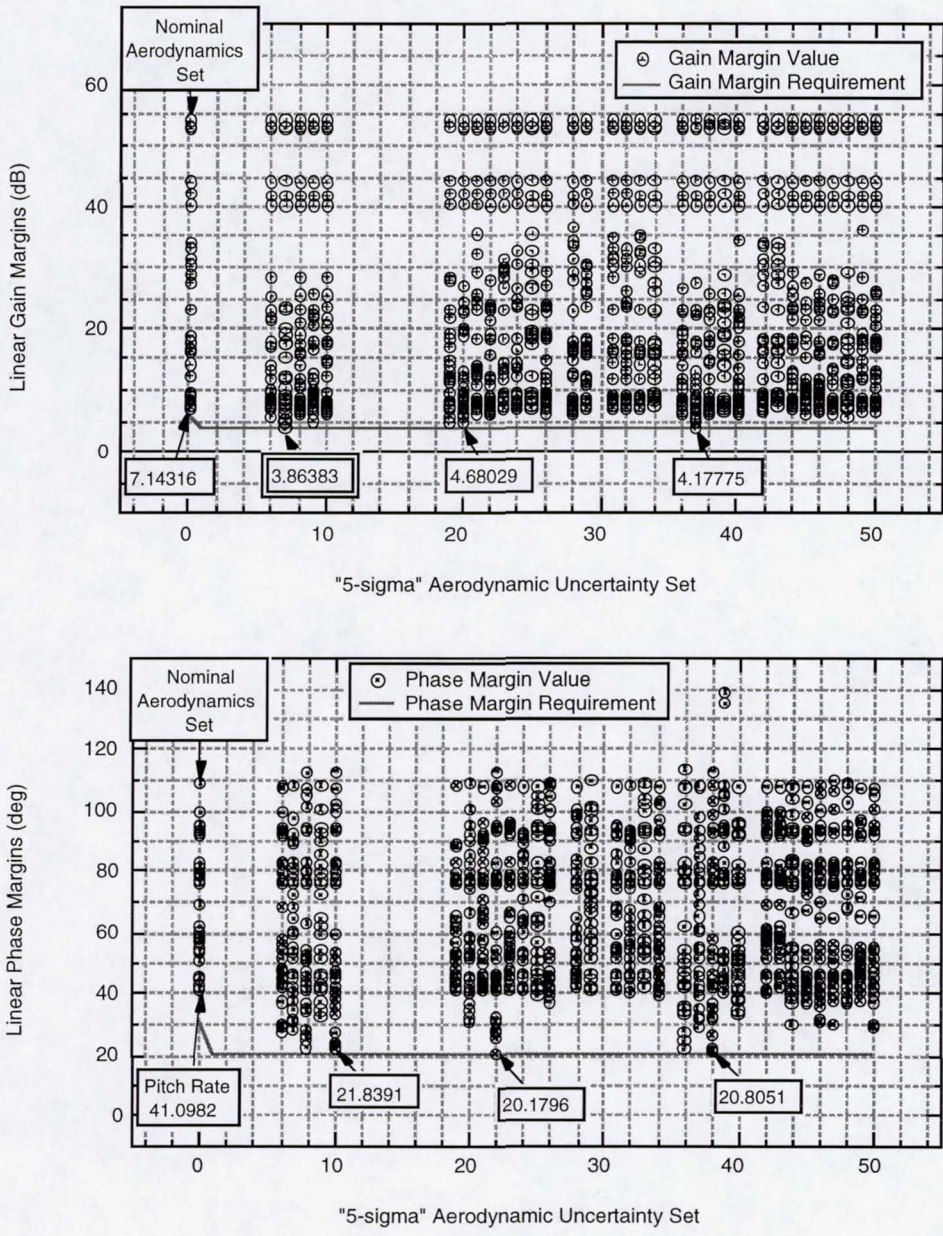


Figure 13: V132 MACH Linear Stability Margins

FLIGHT TEST RESULTS

V132 FF3, the first flight test of MACH, successfully occurred on March 30, 2000, at DFRC. V132 was dropped from the B-52 bomber at an altitude of 38,828 ft, a dynamic pressure of 125 psf and $M = 0.6$. The FCS was engaged at 0.32 seconds after B-52 release and terminated at drogue chute reposition approximately 50 seconds later. The 50 seconds of MACH controlled flight included 44 seconds of free flight and 6 seconds under the drogue chute. During the free flight, the pre-programmed, open-loop guidance commanded angle-of-attack through a profile of 12° , 15° and 11° while bank angle was commanded to zero. Three PTI test objectives were carried out, applying a double sweep of the rudders and ailerons at different angles-of-attack, then performing a bank PTI to study roll performance. The PTI data was used to better characterize the aerodynamics of the surfaces and vehicle. Atmospheric conditions were light-to-moderate turbulence with a 45 knot cross wind (starboard side) relative to the velocity at drop altitude. According to balloon data, the vehicle encountered a head/tail wind shear of about 40 knots midway through the flight during the angle-of-attack maneuver. Flight reconstruction showed that the vehicle flew within its $3\text{-}\sigma$ dispersion envelope and MACH proved to be a DI FCS capable of autonomously controlling an atmospheric lifting body.

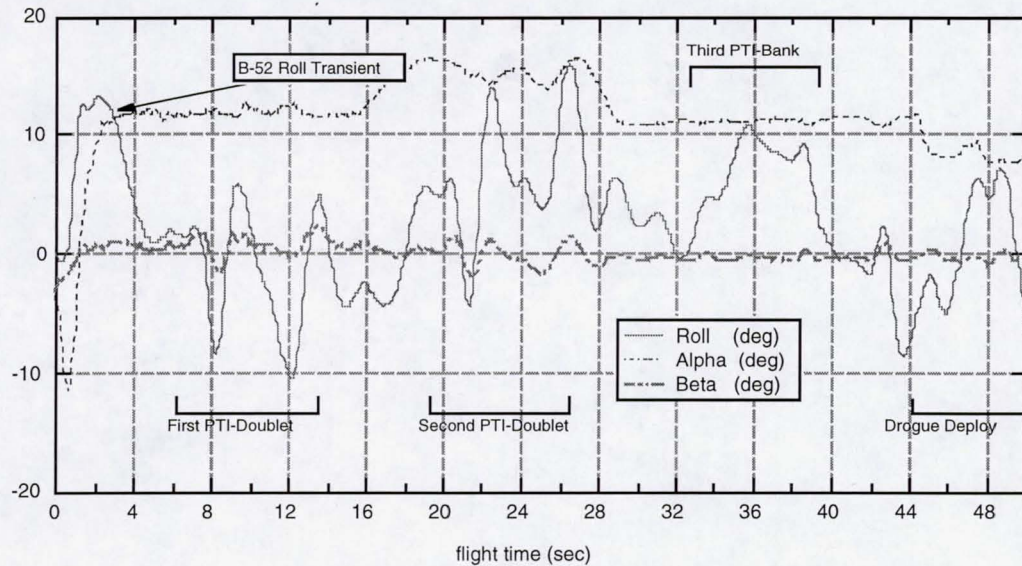


Figure 14: V132 FF3 Vehicle Attitude

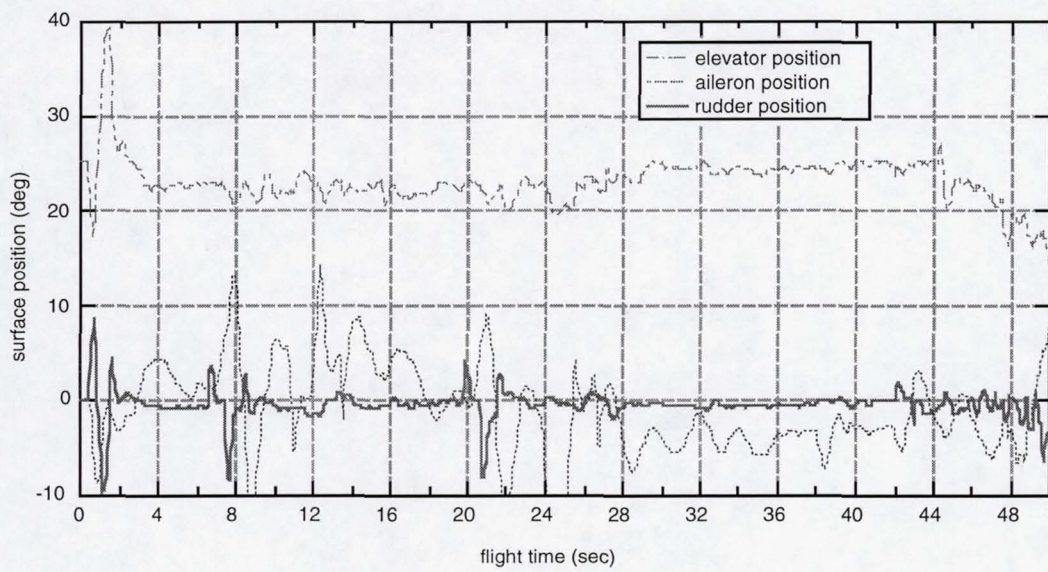


Figure 15: V132 FF3 Surface Positions

CONCLUSIONS

V132 presented many challenges for FCS design. Besides the inherently coupled dynamics and reduced controllability and stability of a subsonic lifting body, a primary design driver was the aerodynamic transient encountered when dropping away from the B-52. Proper design of the MACH FCS helped make the maximum roll and sideslip angles due to B-52 interference in FF3 the lowest of all V132 free flights, despite FF3 being the fastest and highest of the three V132 flight tests. The MACH FCS design effort was able to meet robust stability requirements with a rapidly changing aerodynamic database—partly due to the “automatic gain scheduling” provided by the OBAC—that included relatively large uncertainties.

MACH successfully flew as an autonomous FCS in V132 FF3, the first flight test of this Honeywell implementation of the DI design technique. While MACH may have greater computational requirements than traditional, linear FCS design methods, the least-squares simplification of the aerodynamic database for the OBAC and the general streamlining of MACH source code enabled it to fit well within the computational budget of the V132 flight computer. Though DI is often considered primarily a high-performance control law design method, proper selection of outer loop gains and aerodynamic uncertainty set for the inner loop OBAC enabled MACH to meet all robust stability requirements. Despite being a nonlinear controller, MACH was linearized in order to take advantage of SISO and MIMO linear stability analyses. This MACH design was certified for V132 FF3 with the same linear and nonlinear analyses previously used for the classical, Shuttle-derived FCS in FF1 and FF2, i.e., the certification process was independent of the control law algorithm. The application of MACH to V132 advanced DI FCS technology, thus helping to fulfill the “technology demonstration” goal of the X-38 Project.

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NOTATION

α	alpha or angle of attack
β	beta or angle of sideslip
$C_n\beta$	yawing moment coefficient due to beta
$C_l\beta$	rolling moment coefficient due to beta
$C_l\delta_a$	rolling moment coefficient due to aileron
$C_l\delta_r$	rolling moment coefficient due to rudder
$C_n\delta_a$	yawing moment coefficient due to aileron
$C_n\delta_r$	yawing moment coefficient due to rudder

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