imide, bis-perfluoro-*n*-octylsulfonic acid imide (C8-C8 imide), or perfluoro-*n*butyl-perfluoro-*n*-octylsulfonic acid imide. The experiments involved electrooxidation of methanol on each anode installed as one of the electrodes in a threeelectrode electrochemical test cells. The performances of the electrodes were characterized by galvanostatic polarization measurements and cyclic voltammetry. Of the compounds investigated, C12 acid and C8-C8 imide were found to afford the greatest increases in rates of oxidation of methanol and the greatest reductions in levels of polarization (see figure), relative to those of NafionTM-H.

This work was done by G. K. Surya Prakash, Qun-Jie Wang, and George A. Olah of the University of Southern California and Marshall C. Smart, Sekharipuram Narayanan, and Subbarao Surampudi, of Callech for NASA's Jet Propulsion Laboratory. Further information is contained in a TSP (see page 1).

In accordance with Public Law 96-517, the contractor has elected to retain title to this invention. Inquiries concerning rights for its commercial use should be addressed to:

Innovative Technology Assets Management JPL

Mail Stop 202-233 4800 Oak Grove Drive Pasadena, CA 91109-8099 (818) 354-2240 E-mail: iaoffice@jpl.nasa.gov Refer to NPO-40503, volume and number of this NASA Tech Briefs issue, and the page number

Advanced Ablative Insulators and Methods of Making Them

Reinforced, filled silicones and carbon phenolics are laser-milled to final shapes.

Lyndon B. Johnson Space Center, Houston, Texas

Advanced ablative (more specifically, charring) materials that provide temporary protection against high temperatures, and advanced methods of designing and manufacturing insulators based on these materials, are undergoing development. These materials and methods were conceived in an effort to replace the traditional thermal-protection systems (TPSs) of re-entry spacecraft with robust, lightweight, betterperforming TPSs that can be designed and manufactured more rapidly and at lower cost. These materials and methods could also be used to make improved TPSs for general aerospace, military, and industrial applications.

The ablative materials belong to two families. One family comprises filled,

fiber-reinforced elastomeric carbon phenolics with mass densities that range from 18 to 40 lbm/ft³ (288 to 641 kg/m^3); these materials are designed to protect against heating rates up to about 1,300 Btu/(ft²s) [≈ 15 MW/m²]. The other family comprises filled, fiberreinforced silicones with mass densities that range from 12 to 50 lbm/ft^3 [192 to 800 kg/m³]; these materials are designed to protect against heating rates from 5 to about 400 Btu/(ft²s) [about 0.06 to about 4.5 MW/m²]. The fillers in these materials help to minimize their mass densities, while the fibers help to maximize their strengths.

Design and manufacture of TPSs according to the present approach involve the use of computer-aided design and computer-aided manufacturing (CAD/CAM) methods, including computer numerically controlled (CNC) laser milling. This approach eliminates the labor-intensive steps of machining, fitting, and trimming heat-shield parts in the prior approach to manufacturing. In the present approach, molded panels of the ablative materials are CNC-laser-milled to precise final sizes and shapes and are thus made ready for bonding to heat-shield structures.

This work was done by William M. Congdon of Applied Research Associates, Inc., for Johnson Space Center. For further information, contact the Johnson Technology Transfer Office at (281) 483-3809. MSC-23141

PETIs as High-Temperature Resin-Transfer-Molding Materials PETI-matrix/carbon-fiber composites made by resin-transfer molding have excellent properties.

Langley Research Center, Hampton, Virginia

Compositions of, and processes for fabricating, high-temperature composite materials from phenylethynyl-terminated imide (PETI) oligomers by resin-transfer molding (RTM) and resin infusion have been developed. Composites having a combination of excellent mechanical properties and long-term high-temperature stability have been readily fabricated. These materials are particularly useful for the fabrication of high-temperature structures for jet-engine components, structural components on highspeed aircraft, spacecraft, and missiles. Phenylethynyl-terminated amide acid oligomers that are precursors of PETI oligomers are easily made through the reaction of a mixture of aromatic diamines with aromatic dianhydrides at high stoichiometric offsets and 4-phenylethynylphthalic anhydride (PEPA) as an end-capper in a polar solvent such as N-methylpyrrolidinone (NMP). These oligomers are subsequently cyclodehydrated — for example, by heating the solution in the presence of toluene to remove the water by azeotropic distillation to form low-molecular-weight imide oligomers. More precisely, what is obtained is a mixture of PETI oligomeric species, spanning a range of molecular weights, that exhibits a stable melt viscosity of less than approximately 60 poise (and generally less than 10 poise) at a temperature below 300 °C. After curing of the oligomers at a temperature of 371 °C, the resulting polymer can have a glass-transition temperature (T_g) as high as 375 °C, the exact value depending on the compositions.

As an example, one PETI oligomer, denoted PETI-330, was synthesized as