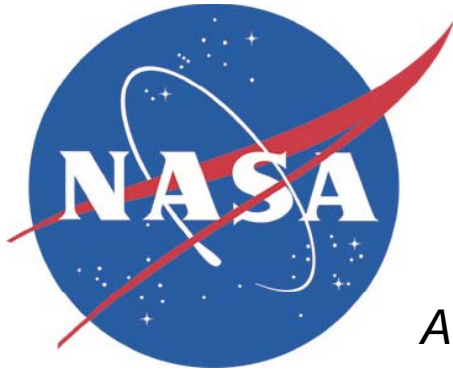

Single and Multi-Pulse Low-Energy Conical Theta Pinch Inductive Pulsed Plasma Thruster Performance

60th JANNAF Propulsion Meeting / 9th MSS / 7th LPS/ 6th SPS
Joint Subcommittee Meeting

JANNAF-2013-3070



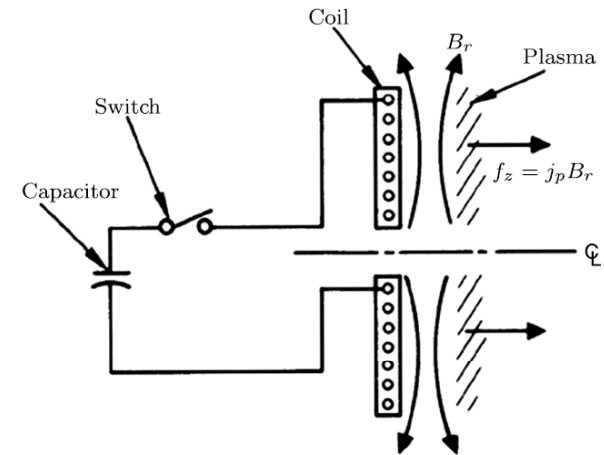
Ashley Hallock
Yetispace, Inc.

Adam Martin, Kurt Polzin, Adam Kimberlin, Richard Eskridge
NASA-Marshall Space Flight Center



Inductive Pulsed Plasma Thrusters

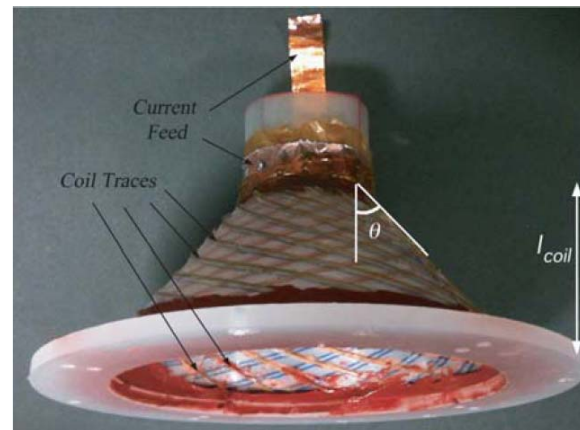
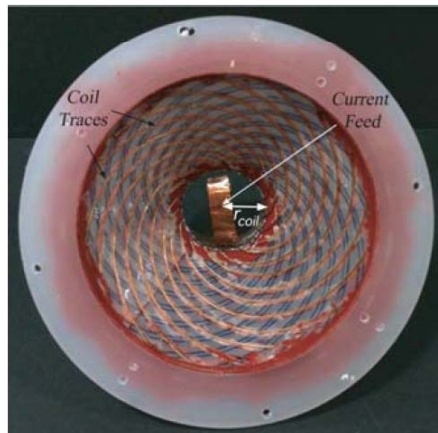
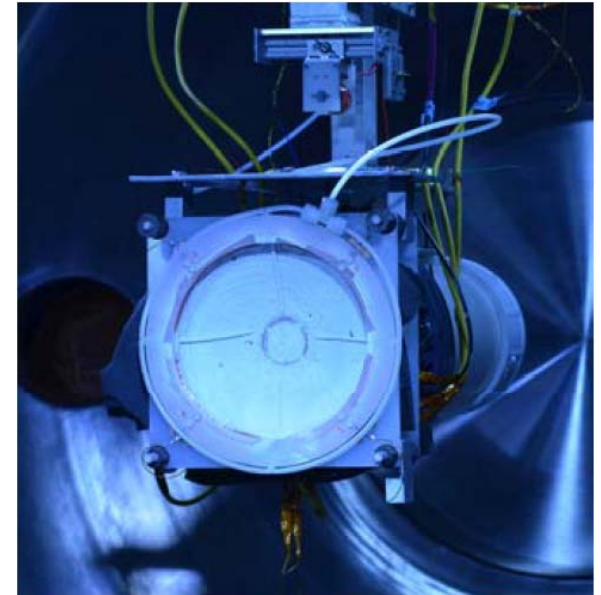
- Energy stored in capacitor banks
- High current switch permit discharge through an inductive coil
- Fast-rising current ionizes/electromagnetically accelerates gas
- Demonstrated and potential benefits
 - Electrodeless
 - Potential to use a wide variety of propellants (Ammonia, CO₂, H₂O, etc.)
 - Constant I_{sp} and thrust efficiency over a wide range of power
 - Regime of relative constant efficiency over a range of I_{sp}
 - Potential to process high power in single thruster (high rep rate)





Conical Theta-Pinch (CTP) IPPT

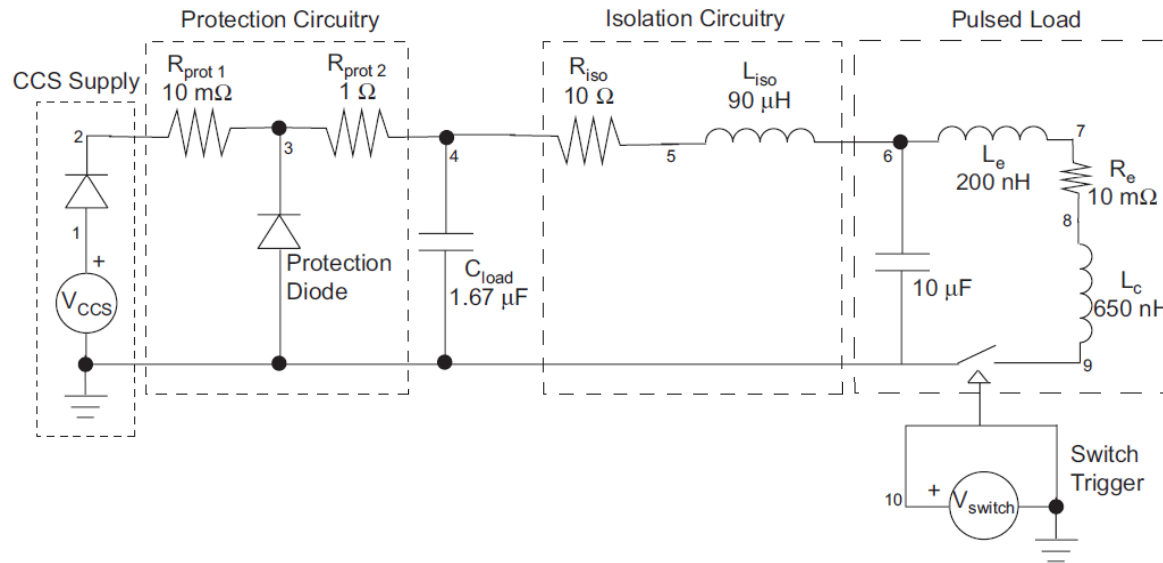
- Propellant potentially more contained and uniform on coil surface
- Three coils fabricated ($\theta=20^\circ, 38^\circ, 60^\circ$) (~ 240 nH)
- Capacitors located directly behind coil (in pressurized enclosure)
- Spark gap-switched capacitor bank
- Direct thrust stand impulse bit measurement



θ	r_{coil}	I_{coil}
20°	4 cm	10 cm
38°	4 cm	10 cm
60°	4 cm	5 cm



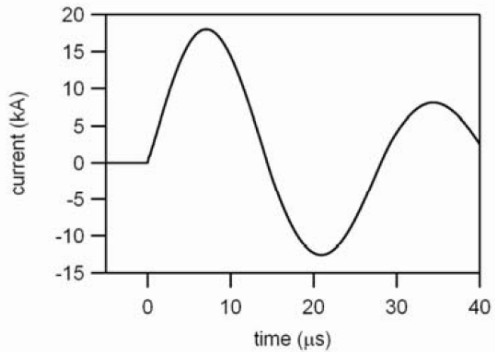
Capacitor Charging System



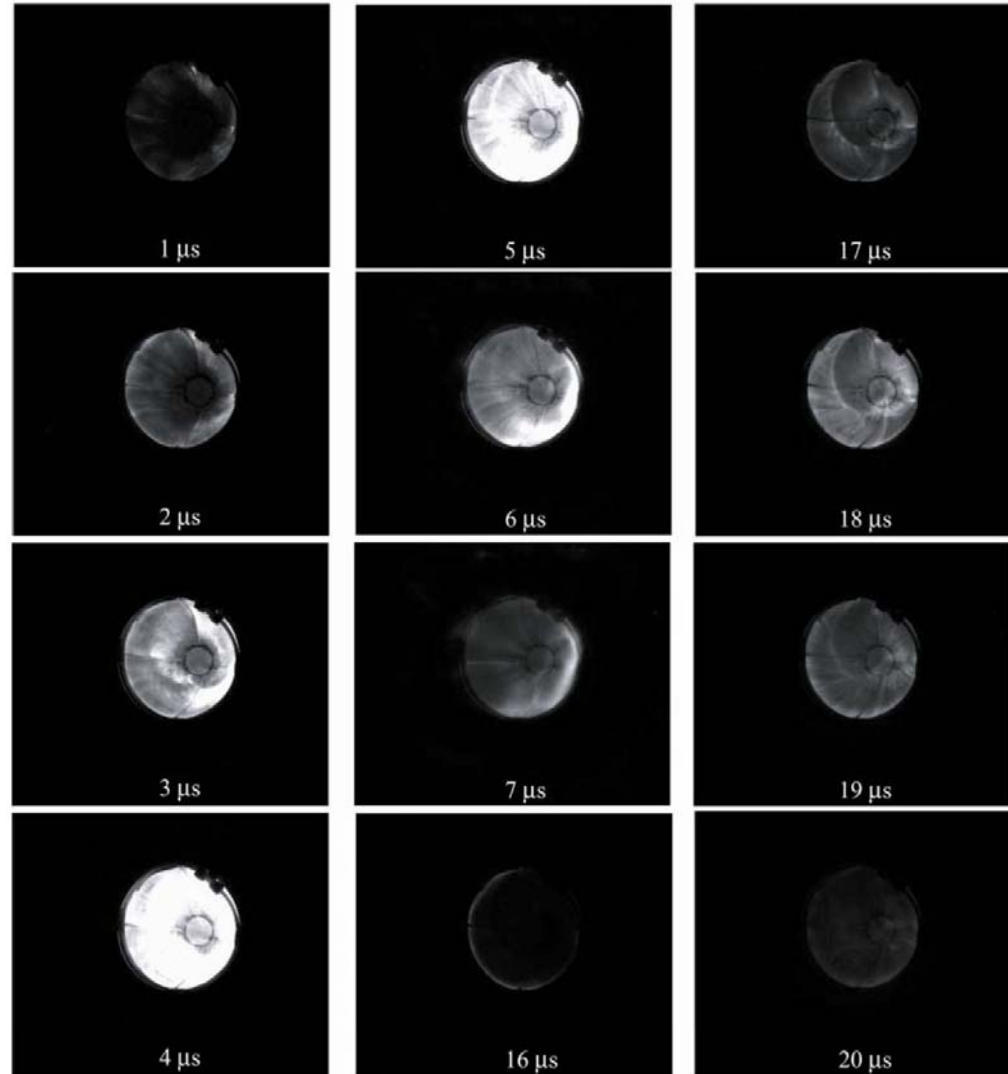
- 40 μF capacitor bank
- 16 kJ/s / 40 kV capacitor charging supply (approximate linear power derating with charge voltage)
- Capacitor bank connected to power supply during pulse – necessitates isolation and protection circuitry
- Pushes repetition-rate limit to ability to rapidly trigger spark-gap switch



High-Speed Imaging

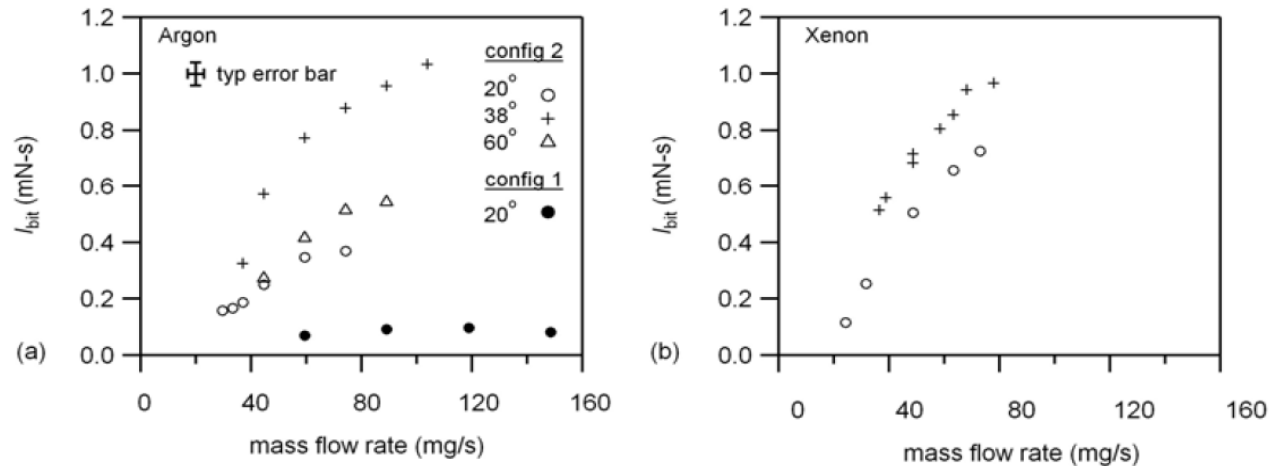


- All light (B&W), 125 ns exposures
- Glow begins at front of thruster and grows backwards at start of 1st/2nd half-cycles
- High intensity over coil in first half-cycle ; lower in second half-cycle
- Visible non-uniformities
- Lower coil current / lower level of gas(?) in 2nd half-cycle





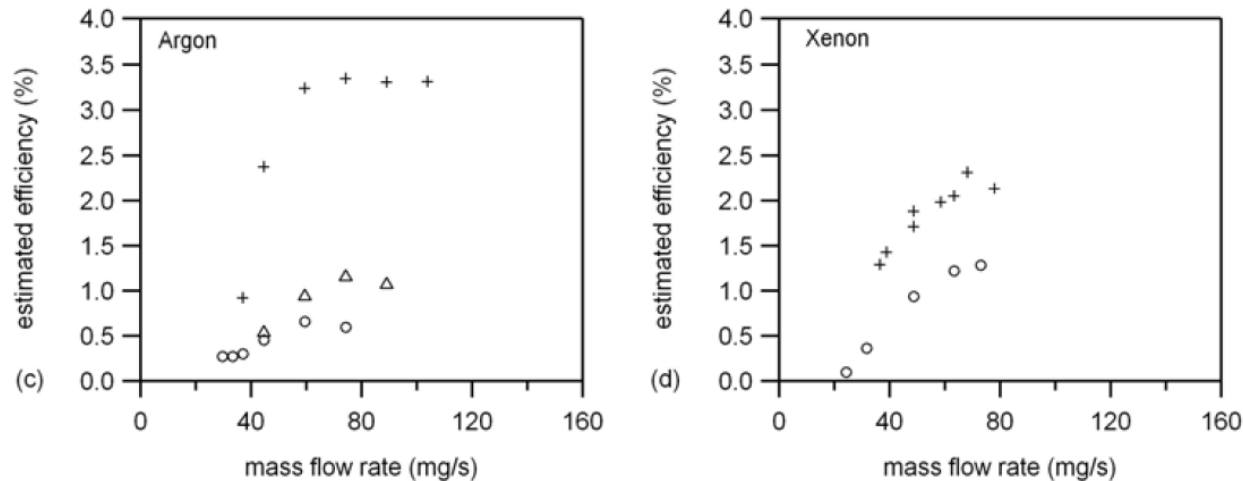
Single-Pulse Performance



- Max I_{bit} of ~1 mN-s
- Max I_{bit} with $\theta=38^\circ$
- Impulse bit peak faster for xenon
 - propellant utilization/more mass near coil?
- High-voltage stand-off issues prevent measurements above flow rates shown



Single-Pulse Performance



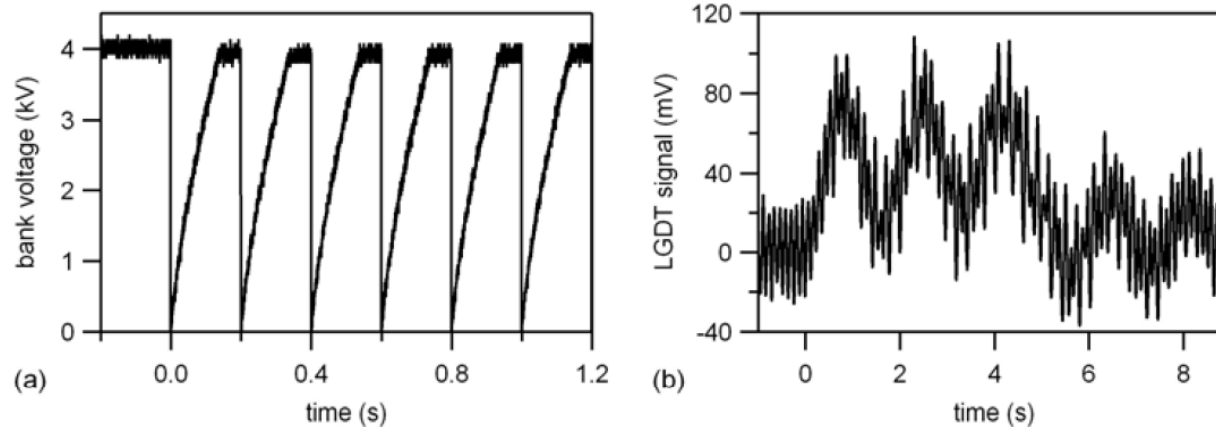
- Steady-state mass flow necessitates estimating efficiency

$$t_{char} = \frac{l}{a} \quad m_{bit} = t_{char} \dot{m} \quad \eta = \frac{I_{bit}^2}{m_{bit} CV_0^2}$$

- Efficiency on argon higher, but both are low
 - Force vector in CTP partially in wrong direction for thrust
 - Similar to peak values in 20-cm PIT (static-fill in late 1960s)
 - Profile/entrainment losses high w/out pulsed injection
 - PIT MkI / MkV (on argon) efficiencies only 15-30% at high energy per pulse



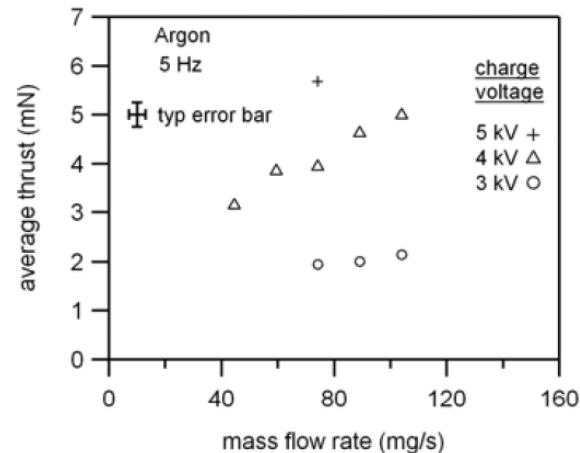
Repetitive Charging and Pulsing



- Repetition rate operation at 5 Hz (up to 2.5 kW average power)
- Repetition rate limit was trigger module for spark gap switch
- Pulsing over 5 seconds
- Thrust stand average displacement yields average thrust during operation



Repetition-Rate Performance



- Average power of 0.9, 1.6, and 2.5 kW (all at 5 Hz)
- 5 kV data in repetition-rate mode greater than 5x the impulse bit in single pulse mode
- To our knowledge, the highest power repetitively-pulsed (i.e. non-CW) discharge
 - Comparison w/ EO-1 PPT (56-70 W @ 1 Hz in ground testing, 12.6 W @ 1 Hz for in-space pulsing)



Conclusions

- Fabricated and tested CTP IPPTs at cone angles of 20°, 38°, and 60°, and performed direct single-pulse impulse bit measurements with continuous gas flow
- Single pulse performance highest for 38° angle with impulse bit of ~1 mN-s for both argon and xenon
- Estimated efficiencies low, but not unexpectedly so based on historical data trends and the direction of the force vector in the CTP
- Capacitor charging system assembled to provide rapid recharging of capacitor bank, permitting repetition-rate operation
- IPPT operated at repetition-rate of 5 Hz, at maximum average power of 2.5 kW, representing to our knowledge the highest average power for a repetitively-pulsed thruster
- Average thrust in repetition-rate mode (at 5 kV, 75 sccm argon) was greater than simply multiplying the single-pulse impulse bit and the repetition rate



Acknowledgements

- MSFC management: Jim Martin, Patrick McRight, Tom Williams, Mary Beth Koelbl, and Tom Brown
- Work benefited from many technical conversations with J. Boise Pearson and Mike LaPointe (MSFC) and Gregory Emsellem (The Elwing Co.)
- High-speed imaging camera loaned to us by Andy Fitchum
- MSFC technical support staff: Tommy Reid, Doug Galloway, Keith Chavers, David Wilkie, Roger Harper, Stan McDonald, and Mark Black
- NASA interns: Kevin Perdue, Alexandra Toftul, Andrea Wong, Kevin Bonds, and Mark Becnel

- Work funded by the In Space Propulsion Project of the Game-Changing Division (GCD) of NASA's Office of the Chief Technologist. The GCD principle investigator was Chuck Taylor, and the project manager was Tim Smith.