

# Mars Ascent Vehicle Design for Human Exploration

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**In NASA's evolvable Mars campaign, transportation architectures for human missions to Mars rely on a combination of solar electric propulsion and chemical propulsion systems. Minimizing the Mars ascent vehicle (MAV) mass is critical in reducing the overall lander mass and also eases the requirements placed on the transportation stages. This paper presents the results of a conceptual design study to obtain a minimal MAV configuration, including subsystem designs and mass summaries.**

## Nomenclature

<i>ARV</i>	= Asteroid Redirect Vehicle
<i>BOL</i>	= Beginning of life
<i>CBE</i>	= Current Best Estimate
<i>DOI</i>	= Descent Orbit Insertion
<i>ECLS</i>	= Environmental Control and Life Support
<i>EVA</i>	= Extravehicular Activity
<i>FAS</i>	= Flight Analysis Software
<i>ISRU</i>	= In-Situ Resource Utilization
<i>IVA</i>	= Intravehicular Activity
<i>Isp</i>	= engine specific impulse
<i>K</i>	= Kelvin
<i>kg</i>	= kilogram
<i>kN</i>	= kilonewton
<i>kPa</i>	= kilopascals
<i>kW</i>	= kilowatt
<i>LCH<sub>4</sub></i>	= Liquid methane
<i>LCI</i>	= Layered Composite Insulation
<i>LEA</i>	= Launch, Entry, and Abort
<i>LOX</i>	= Liquid Oxygen
<i>LTP</i>	= Launch Targeting Processor
<i>m</i>	= meter
<i>m<sup>2</sup></i>	= square meters
<i>m<sup>3</sup></i>	= cubic meters
<i>MAG</i>	= Maximum Absorbency Garment
<i>MAV</i>	= Mars Ascent Vehicle
<i>MDM</i>	= Mars Descent Module
<i>MLI</i>	= Multi-Layer Insulation
<i>MOI</i>	= Mars Orbit Insertion

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<i>MPa</i>	=	Megapascals
<i>MPS</i>	=	Main Propulsion System
<i>NDS</i>	=	NASA Docking System
<i>OMP</i>	=	Orbital Maneuvering Processor
<i>psi</i>	=	pounds per square inch
<i>POST</i>	=	Program to Optimize Space Trajectories
<i>RCS</i>	=	Reaction Control System
<i>SEP</i>	=	Solar Electric Propulsion
<i>SLS</i>	=	Space Launch System
<i>sol</i>	=	Mars solar day
<i>t</i>	=	metric ton
<i>TCS</i>	=	Thermal Control System
<i>TEI</i>	=	Trans-Earth Injection
<i>V</i>	=	Volt
<i>VJ</i>	=	Vacuum Jackets
<i>VME</i>	=	Versa Module Europa
<i>W</i>	=	Watt
<i>WMS</i>	=	Waste Management System

## I. Introduction

THE NASA Human Spaceflight Architectures Team is currently studying an Evolvable Mars Campaign<sup>1</sup> that aims to define an evolutionary path from current spaceflight infrastructure and capabilities to the ultimate goal of landing people on the surface of Mars and returning them to Earth. The Mars ascent vehicle (MAV) design has a significant effect on the Mars transportation architecture. Changes in MAV mass estimates ripple back through the architecture, affecting the entry and descent stage design of the lander as well as the performance for Earth launch and transportation to Mars.

Preliminary analyses early in 2014 indicated that the minimum Mars lander mass capable of supporting human missions was 43 t at Mars entry. This minimum mass was determined by the single largest indivisible payload element required for the surface mission, the Mars ascent vehicle. At that time, the MAV (along with the ISRU plant, small rover and tunnel) was estimated to be 18 t delivered to the surface and 40 t fully loaded with liquid oxygen gathered from the Martian atmosphere. Transportation studies at that time indicated that a solar electric propulsion (SEP) vehicle derived from the Asteroid Redirect Vehicle (ARV), and limited to 150 kW beginning of life (BOL) power generation, could deliver 40 t to Mars orbit<sup>2,3</sup>. With a potential 3 t gap in performance needed for delivering Mars landers, effort began to further refine the SEP trajectory and performance analysis to increase performance and further refine Mars lander mass estimates.

In the fall of 2014, a conceptual design study was initiated to refine mass estimates for a Mars ascent vehicle capable of lifting 4 astronauts off the surface of Mars. Results of that study are presented in this paper including operations and functional requirements, configuration, and mass properties as well as subsystem design and vehicle performance sensitivities.

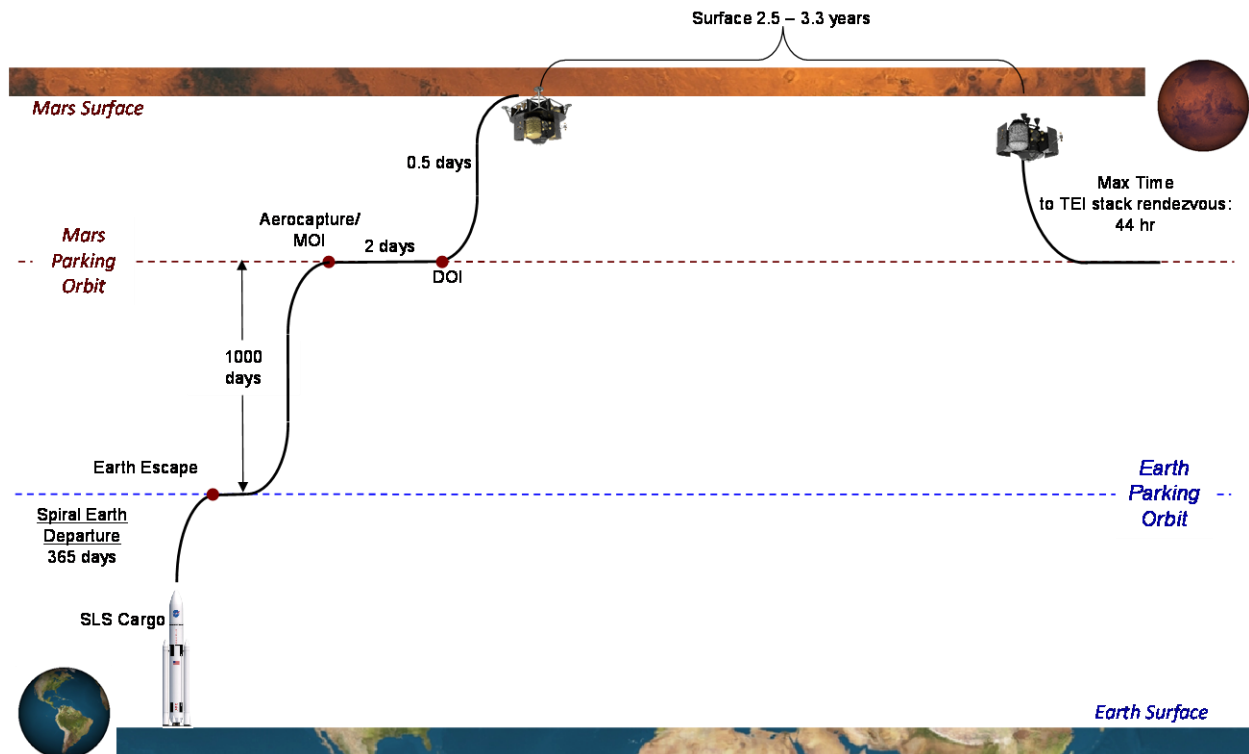
## II. Operations

The MAV's primary objective is to lift crew and cargo off the surface of Mars and dock with an orbiting Mars-Earth transportation vehicle. The MAV (without the oxygen propellant) is predeployed years in advance of a crew landing to allow adequate time for propellant generation. The MAV uses oxygen that is collected and liquefied on the Martian surface along with methane brought from Earth as propellant.

Transit time from Earth launch to arrival in Mars orbit ranges between 300 days and 1400 days depending on the transportation system used. Purely SEP systems take the longest when power limited to 150 kW, while purely chemical transportation systems follow minimum energy trajectories which are much shorter in duration than those involving SEP propulsion. Hybrid SEP/chemical systems fall between that range<sup>4,5</sup>. The baseline assumed for this study is an SEP transit with flight times given in Figure 1. During transit, the MAV relies on the descent stage of the lander for power, communications, and thermal control.

Shortly after arrival into a 250 km x 33,800 km Mars orbit, commonly referred to as a 1 sol orbit, the lander carrying the MAV detaches from the Earth-Mars transportation system. After a brief period of final checkouts and phasing to align with the targeted landing site, the lander descends to the surface. Once on the surface the lander must be connected to a surface power generator, currently assumed to be a fission power source that is delivered

with the MAV or on an earlier lander. Connection to surface power is assumed to occur within 24 hours after landing. Surface power is required for liquid oxygen production and to prevent boiloff of the ascent propellants. Oxygen production rates are dependent on available power, but the production duration is expected to be up to one year. During surface operations the MAV relies on the descent stage for communications, thermal control, and connection to surface power.



**Figure 1. MAV Mission Overview.**

To minimize requirements on the MAV, as well as mass and complexity, crew access is limited during the surface missions to brief periods of checkout operation before ascent<sup>6</sup>. The MAV is not intended to be a secondary habitat, though it is capable of sheltering the crew for a limited period of time in an emergency situation. Nominal crew access begins the day before launch. Two crew members travel to the MAV from their surface habitat via pressurized rover and perform system checkout operations and stow cargo in the MAV. The remaining crew members arrive the day of the launch. Crew ingress is facilitated by a pressurized tunnel connected to the rover. The tunnel minimizes the potential to contaminate the MAV with martian soil which would become airborne once the MAV reached orbit. It also allows the crew to leave surface exploration suits behind in the rover and enter the MAV and ascend in the lighter and more flexible launch and entry suits. One key assumption is that the crew remain in their suits with visors up during ascent through rendezvous and dock except for brief hygiene breaks. This minimizes the “feed the leak” duration and consumables penalty in the event of a cabin leak and minimizes required cabin size by eliminating the need to store 4 empty suits.

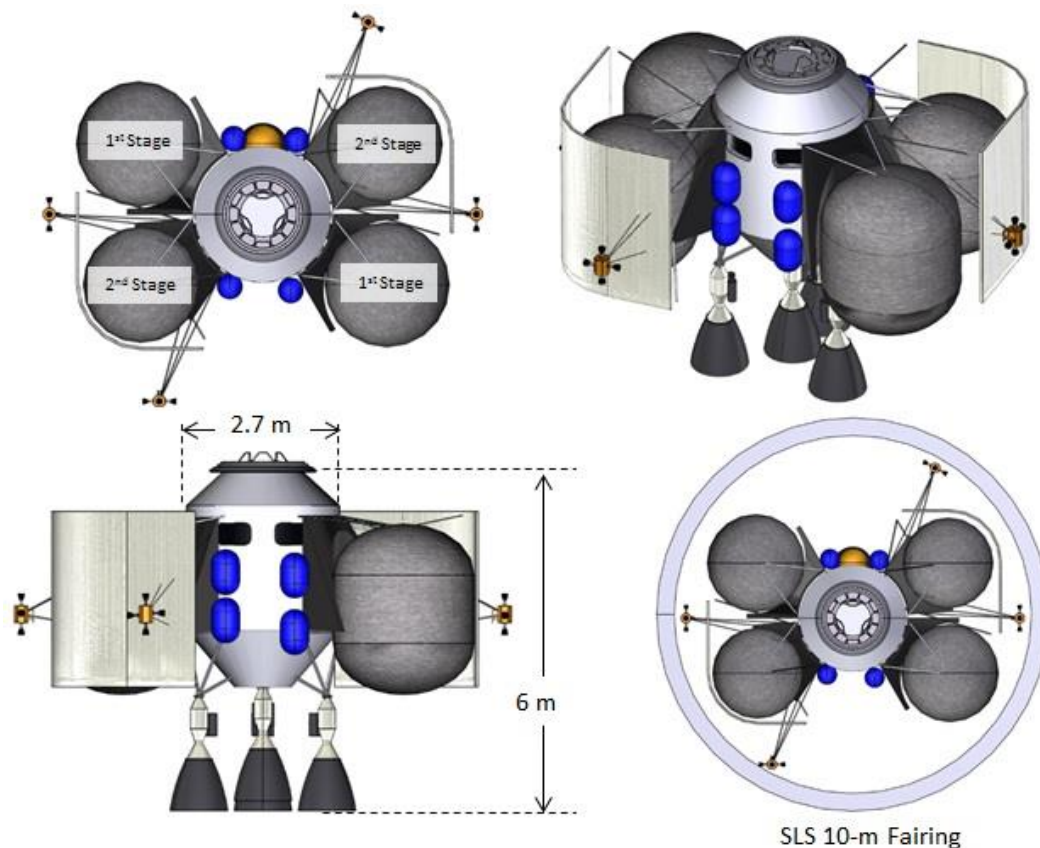
Just prior to ascent all support services from the descent stage are discontinued, and the MAV becomes self-sufficient. A powered ascent leaves the MAV in a 100 x 250 km altitude orbit. The MAV then circularizes into a 250 km orbit and awaits optimum phasing for rendezvous with the Earth return vehicle (in the 1 sol parking orbit). Once docking is achieved and crew and cargo are transferred, the MAV detaches and performs a final disposal maneuver into an orbit that will not interfere with future Mars orbit operations.

### III. Vehicle Overview

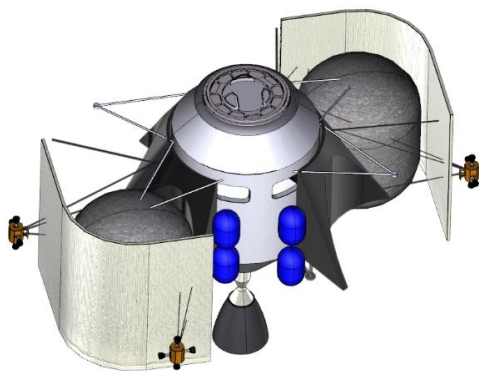
#### A. Configuration

The MAV is a two-stage vehicle with three 100 kN engines on the first stage and a single engine (same thrust) on the second stage. The first stage consists of only a main propulsion system (MPS) with two sets of nested tanks, for liquid oxygen and liquid methane. These components are dropped after the first stage burn, leaving the second stage which contains the cabin, crew, propulsion system (MPS and RCS), along with the supporting subsystems,

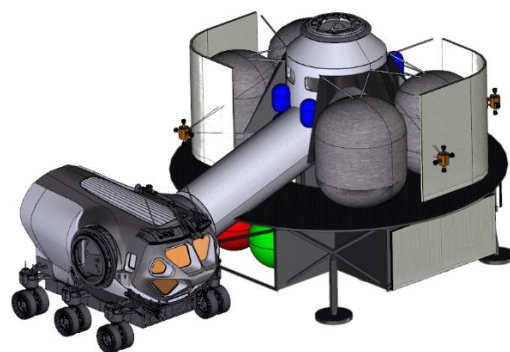
including fixed thermal radiators (wrapped around the second stage tanks) and the fuel-cell power generation system. Like the first stage, the second stage has two sets of nested propellant tanks. Figure 2 shows both stages and packaging in a 10-m diameter SLS fairing, while Figure 3 displays only the second stage. The MAV height is 5.8 m with a cabin diameter of 2.7 m, which provides an internal volume of 17.5 m<sup>3</sup>. Two hatches allow for entry from the Mars surface and docking to the Earth return vehicle. The NASA docking system<sup>7</sup> is assumed for the top hatch, and a 40 inch rectangular hatch is assumed on the side. Figure 4 shows the MAV on the surface connected to the rover via pressurized tunnel. This arrangement of cabin and propulsion system components allows for relatively straightforward crew access as well as a low center of gravity for the overall lander, which is desired for controllability during the entry, descent and landing phases.



**Figure 2. MAV First and Second Stages.**



**Figure 3. MAV Second Stage.**



**Figure 4. Crew Access Tunnel to MAV.**

Figure 5 shows the nested tank information for both stages. Each first stage tank contains fuel and oxidizer masses of 2,298 kg and 7,841 kg, respectively. This gives a total first stage propellant load of 20.3 t. The second stage tank is slightly smaller and contains 1191 kg of LCH<sub>4</sub> and 3523 kg of LOX, giving a combined propellant load of 9.4 t for both tanks.

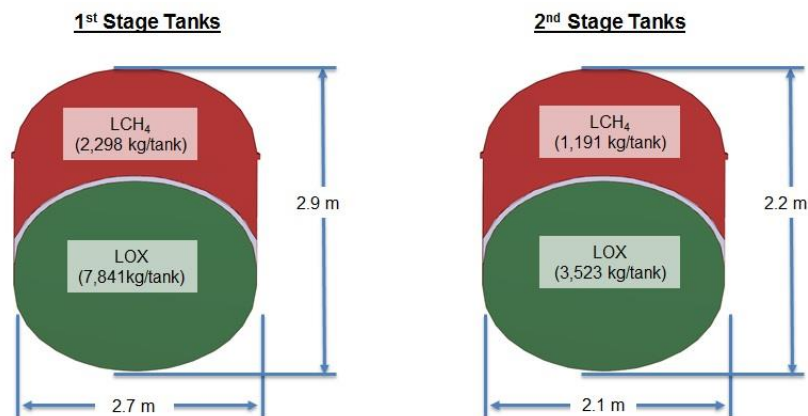


Figure 5. Nested Tank Information.

## B. Vehicle Mass Properties

The MAV subsystem masses, including the totals, are shown in Table 1. The total MAV mass at the beginning of the crewed ascent is 39.1 t, which consists of a 3.8 t crew cabin, 23.5 t first stage, and 11.8 t second stage. The listed propellants represent the total loaded masses, which contain 19.4 t of usable propellant on the first stage and 8.7 t on the second stage. Note: all propulsion items (including engines and tanks) are part of the dry masses of the first and second stages.

The total mass at Earth departure is 15.7 t. The MAV is launched without liquid oxygen, which will be provided on the Martian surface by the lander ISRU system. Also absent are the return cargo, crew, and their space suits.

## IV. Vehicle Overview

### A. Trajectory

The ascent performance of the MAV was modeled using Program to Optimize Space Trajectories (POST). The flight originates from 30° north latitude and ends in the initial low Mars orbit of 100 km x 250 km with a 30° inclination. Three engines are used for the first stage, producing a liftoff acceleration of 0.78 Earth g's. Staging occurs at 225.8 seconds with a maximum acceleration of 1.52 Earth g's. The single second stage engine burns for 192.6 seconds to insertion into the low Mars orbit. Maximum acceleration during this burn is 1.10 Earth g's. These stage burn durations are optimized by POST based on the input propellant mass fractions of the stages.

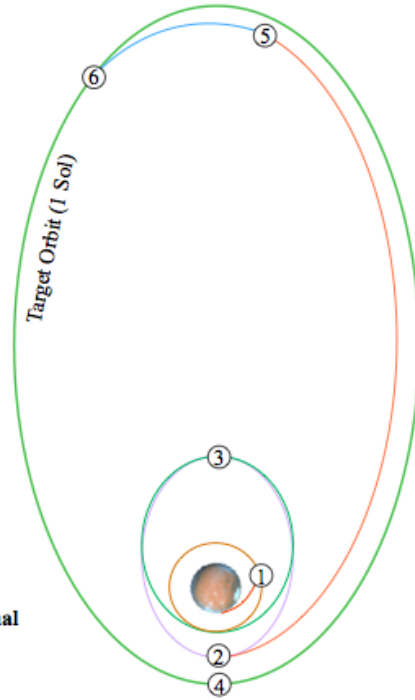
The MAV then starts the sequence of burns using either the main engine of the 2nd stage or RCS, whichever is appropriate for the burn, to rendezvous with the Trans-Earth Habitat in a 1 sol orbit. The rendezvous analysis was performed by Jeffrey Gutkowski, NASA Johnson Space Center, using Flight Analysis Software (FAS) specifically the Launch Targeting Processor (LTP) and Orbital Maneuvering Processor (OMP). The results are shown in Fig. 6. The maximum acceleration during this phase of the mission is about 1.4 g's and occurs at the end of the "Coelliptic" maneuver (the only main engine burn after the 100 km x 250 km insertion).

Table 1. MAV Masses.

Subsystem	Mass (kg)	
	Earth Departure	Mars Liftoff
<b>Crew Cabin</b>	<b>3,151</b>	<b>3,846</b>
Structures	858	858
Power	377	377
Avionics	407	407
Thermal	542	542
ECLS	298	298
Cargo	411	1,106
Non-Prop. Fluids	258	258
<b>1st Stage</b>	<b>7,791</b>	<b>23,473</b>
Dry Mass	3,195	3,195
LOX	0	15,681
LCH4	4,597	4,597
<b>2nd Stage</b>	<b>4,719</b>	<b>11,756</b>
Dry Mass	2,378	2,378
LOX	0	7,036
LCH4	2,341	2,341
<b>Total Masses:</b>	<b>15,662</b>	<b>39,075</b>

Event	Hp / Ha (km)	$\Delta V$ (m/s)	Time From Launch (hh:mm)
1. Raise periapsis	166 / 250	15	00:57
2. Phasing	162 / 471	50	01:51
3. Height	200 / 470	8	04:43
4. Coelliptic (NSR)	200 / 33,792	1147	24:50
5. Lambert-targeted (TPI)	240 / 33,762	4.6	34:50
6. Braking Maneuver (TPF)	250 / 33,796	1.7	43:55

- Incorporated within the  $\Delta V$  numbers above the
  - Out-of-plane contribution was 4 m/s
  - Radial contribution was 4 m/s
- Therefore the majority of the total  $\Delta V$  comes from in-plane  $\Delta V$ , indicating that each maneuver is performed at its respective optimal location

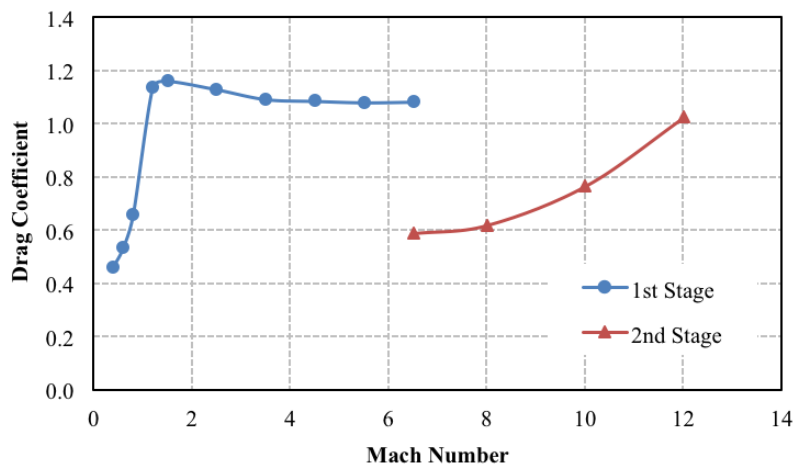


**Figure 6. Phasing and Rendezvous**

Shorter flight durations from the surface to rendezvous with the Earth return vehicle are possible. The option presented here allows for multiple launch opportunities per day every day during the surface mission assuming the Earth return vehicle is positioned appropriately for a co-planar ascent. Reducing flight duration affects the launch window availability and required  $\Delta V$ . For a difference of only 3 additional m/s of  $\Delta V$ , a 24 hour flight to 1 sol can be achieved, but this trajectory solution may only be available a few days out of each month. A 12 hour flight could be possible with an additional 190 m/s of  $\Delta V$ , but the opportunities are likely to be even less frequent, and the launch windows may only be minutes long. For this MAV design, propellant loads are calculated assuming the slightly higher  $\Delta V$  for a 24 hour flight, and the crew consumables are computed for the 44 hour flight. This allows a one day ascent if conditions permit, while protecting for a longer flight if needed.

## B. Aerodynamics

The atmospheric pressure on the surface of Mars is only about 0.6% of Earth's sea level pressure and drops off quickly with altitude, however the effect on ascent vehicle performance cannot be neglected. The drag coefficients used in the POST simulation are shown in Fig. 7 and represent a preliminary assessment of the MAV aerodynamics. FUN3D (Fully Unstructured Navier-Stokes) is used in the analysis with an 8-species Mars atmosphere composition (non-reacting) and the MAV configuration models. The assumed reference length and area are 5.13 m and 22.06 m<sup>2</sup>, respectively. The displayed coefficients are for a zero angle of attack.



**Figure 7. MAV Drag Coefficients.**

### C. Comparison to Apollo and Altair Lunar Ascent Vehicles

The Apollo Lunar Module ascent stage is the only vehicle that has carried humans off the surface of another planetary body. Over six missions these vehicles carried a total of 12 astronauts and 382 kg of samples off the surface of the moon to rendezvous and dock with the orbiting command module. The Altair lunar lander (part of NASA's Constellation Program that was cancelled in 2010) was to have a larger and more capable ascent module. It was designed to carry 4 crew members instead of 2 and a higher allocation for the return of lunar samples on each ascent. While the Altair ascent module never flew, with three years of significant design effort and two years of formulation prior to that, it is the highest fidelity design of an ascent vehicle since the Apollo Lunar Module. In comparison, the Mars Ascent Vehicle is significantly more massive than both lunar ascent vehicles. This is primarily due to the very different propulsion requirements of the Mars vs. Lunar ascent trajectories<sup>8</sup>. Table 2 provides a comparison of requirements and masses.

**Table 2. Comparison to Apollo and Altair Ascent Vehicles.**

	<b>Apollo</b>	<b>Altair</b>	<b>MAV</b>
Number of Crew	2	4	4
Crew Cabin Pressurized Volume (m <sup>3</sup> )	6.65	17.5 (including airlock)	17.5
Ascent to Docking Flight Time (hrs)	2.1-3.7	2.5	24 - 44
Ascent $\Delta V$ (m/s)	1,900	1,985	5,274
Dry Mass (kg)	1,901	2,615	9,419
Propellant Mass (kg)	2,492	3,147	29,655
Total Mass (kg)	4,795	6,190	39,075

## V. Vehicle Systems

### A. Propulsion

The MAV propulsion system is separated into two stages. The first stage contains three 100 kN engines that consume liquid oxygen (oxidizer) and liquid methane (fuel). The engine operates at a specific impulse of 360 s and a propellant oxidizer-to-fuel mixture ratio of 3.5. The second stage contains a single main engine (same engine as first stage) along with four banks of four (16 total) reaction control engines. Each RCS engine produces 445 N of thrust and a specific impulse of 340 s. The corresponding mixture ratio is 3.0. The first stage MPS pulls propellant from two nested tanks on that stage (see Figure 4 above). The second stage tanks provide propellants to both the MPS and RCS engines. RCS propellants must be vaporized and then pumped into accumulator tanks before usage. Gaseous propellants allow high pulse rates during attitude control. The maximum pressure in the accumulator tanks is 27.6 MPa. The gas-fed engine assumption proved to be complex and resulted in a high power requirement. Therefore, liquid-fed RCS engines will be studied for future design iterations.

The tank pressures are maintained at 345 kPa. The LOX is pressurized with helium, stored in ambient bottles at 31 MPa. Each bottle has a blowdown ratio of 3.0, and the process is assumed to be isentropic. The LCH<sub>4</sub> tank pressurization is autogenous. Cryocoolers maintain the propellant temperatures to prevent boiloff. The liquid oxygen is kept at 94 K, while the liquid methane is stored at 112 K. Other assumptions include a 5% ullage volume for each propellant, 1% residual propellants, and a 1% fuel bias. All propulsion masses are listed in Tables 3 and 4 for the first and second stages, respectively.

**Table 3 First Stage Propulsion Masses.**

System	Item	Qty	Unit Mass (kg)	Basic Mass (kg)	Growth (%)	CBE Mass (kg)
<b>Dry Mass</b>						
<b><i>MPS</i></b>				<b>2347</b>	<b>25</b>	<b>2933</b>
	Main Engine	3	181	544	25	680
	Oxidizer Subsystem	1	127	127	25	158
	Fuel Subsystem	1	135	135	25	168
	Pressurization & Pneumatics	1	199	199	25	248
	Mounting Hardware	1	100	100	25	126
	Nested Tanks	2	300	600	25	750
	MPS Tank Cryocoolers/BAC	1	44	44	25	55
	MPS Tank Insulation	1	598	598	25	748
<b>Non-Propelled Fluids</b>						
<b><i>Residuals</i></b>						<b>664</b>
	Oxidizer					479
	Fuel					185
<b><i>Fuel Bias</i></b>						<b>43</b>
	Fuel					43
<b><i>Reserves</i></b>						<b>194</b>
	Oxidizer					151
	Fuel					43
<b><i>Pressurant</i></b>						<b>58</b>
	Gaseous Helium					32
	Autogenous Fuel					25
<b>Used Propellant</b>						
<b><i>Mnvr Prop</i></b>						<b>19,267</b>
	Oxidizer					14,986
	Fuel					4,282
<b><i>Engine Start/Stop</i></b>						<b>85</b>
	Oxidizer					66
	Fuel					19



**Table 4. Second Stage Propulsion Masses.**

System	Item	Qty	Unit Mass (kg)	Basic Mass (kg)	Growth (%)	CBE Mass (kg)
<b>Dry Mass</b>						
<b><i>MPS</i></b>				<b><i>1247</i></b>	<b><i>25</i></b>	<b><i>1559</i></b>
	Main Engine	1	181	181	25	227
	Oxidizer Subsystem	1	50	50	25	62
	Fuel Subsystem	1	51	51	25	64
	Pressurization & Pneumatics	1	143	143	25	178
	Mounting Hardware	1	43	43	25	53
	Nested Tanks	2	138	276	25	345
	MPS Tank Cryocoolers/BAC	1	98	98	25	123
	MPS Tank Insulation	1	370	370	25	463
	Radiators	1	36	36	25	44
<b><i>RCS</i></b>				<b><i>561</i></b>	<b><i>20.45</i></b>	<b><i>676</i></b>
	Heating	1	17	17	25	21
	Heat Exchanger	1	10	10	25	13
	High Pressure Storage	1	54	54	25	68
	Lines & Hardware	1	50	50	18	59
	Propellant Distribution	1	116	116	17	136
	RCS Engine POD (4 engines)	4	72	287	21	347
	Thermal Control	1	18	18	23	23
	Instrumentation	1	9	9	16	10
<b>Non-Propelled Fluids</b>						
<b><i>Residuals</i></b>						<b><i>231</i></b>
	Oxidizer					169
	Fuel					62
<b><i>Fuel Bias</i></b>						<b><i>41</i></b>
	Fuel					41
<b><i>Reserves</i></b>						<b><i>87</i></b>
	Oxidizer					67
	Fuel					19
<b><i>Pressurant</i></b>						<b><i>23</i></b>
	Gaseous Helium					17
	Autogenous Fuel					6
<b>Used Propellant</b>						
<b><i>Mnvr Prop</i></b>						<b><i>8,622</i></b>
	Oxidizer					6,689
	Fuel					1,933
<b><i>Engine Start/Stop</i></b>						<b><i>57</i></b>
	Oxidizer					44
	Fuel					13

**B. Structures**

A detailed finite element analysis is performed to estimate the MAV structures. The model is shown Figure 8, which also includes the driving assumptions. The results are listed in Table 5. The propellant tanks are estimated in a similar manner (given above in Tables 3 and 4). Micrometeoroid/orbital debris shielding is not included. Further study on exposure risk and required shielding is needed.

Launch, Landed Mass	16000	kg
Ascent Mass	38750	kg
Pressurized Volume	2.20	m <sup>3</sup>
Internal Pressure	101	kPa
Cabin Nominal Length	3.8	m
Cabin Nominal Dia	2.7	m
<b><u>Loads:</u></b>		
Launch, axial	+5, -2	g
Launch, lateral	+/-2	g
EDL, axial	4.5	g
EDL, lateral	+/-0.5	g
Ascent, axial	1.3	g
Ascent, lateral	+/-0.5	g
<b><u>Propellants:</u></b>		
LO2, stage 1	15682	kg
LCH4, stage 1	4596	kg
LO2, stage 2	6970	kg
LCH4, stage 2	2074	kg
Bogey for tank masses:	780	kg
<b><u>Engines:</u></b>		
Mass	181	kg
Thrust	100	kN
<b><u>Radiators:</u></b>		
Area	32	m <sup>2</sup>

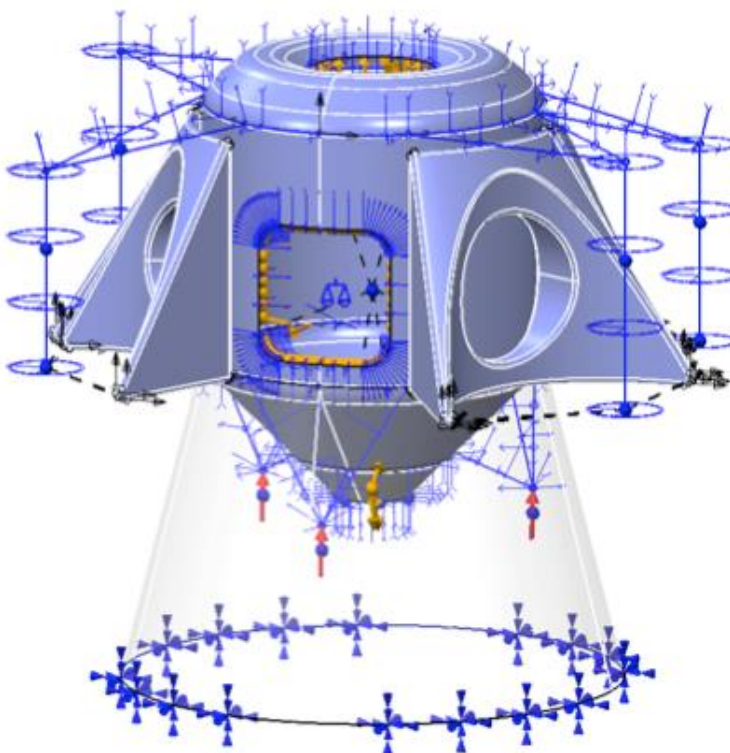


Figure 8. Structures Finite Element Module and Assumptions.

**Table 5. First and Second Stage Structures Masses.**

System	Item	Qty	Unit Mass (kg)	Basic Mass (kg)	Growth (%)	CBE Mass (kg)
<b>First Stage</b>						
<b>Tank Support</b>				<b>163</b>	<b>25</b>	<b>204</b>
	Sandwich Panel Assemblies	2	76	152	25	189
	Strut Assemblies	2	6	12	25	14
<b>Thrust Structure</b>				<b>20</b>	<b>25</b>	<b>25</b>
	Strut Assemblies	3	7	20	25	25
<b>Second Stage</b>						
<b>Cabin</b>				<b>686</b>	<b>25</b>	<b>858</b>
	Composite Sandwich Shell	1	237	237	25	297
	Upper & Lower Ring Frame	1	21	21	25	26
	Hatch Frame	1	11	11	25	14
	Floor	1	50	50	25	63
	Side Hatch	1	100	100	25	125
	Top Hatch (NDS Passive Side)	1	92	92	25	115
	Window Assemblies	2	23	45	25	56
	Secondary Structure	1	68	68	25	85
	Descent Module Interface	1	40	40	25	50
	Engine Support Structure	1	22	22	25	28
<b>Thrust Structure</b>				<b>101</b>	<b>25</b>	<b>126</b>
	Sandwich Panel Assemblies	2	45	90	25	112
	Strut Assemblies	2	6	11	25	14

**C. Power**

The maximum power required for MAV operation is estimated at 10 kW. The power generation system consists of three solid oxide fuel cell power plants, each producing 5 kW. This allows for maximum MAV power usage to be carried by two fuel cells with one fuel cell held in reserve. The fuel cells draw CH<sub>4</sub> and O<sub>2</sub> from the second stage propellant tanks. Solid oxide cells are used due to their higher power densities and efficiencies. The reactant amounts are computed based on a 60% fuel cell efficiency with an applied margin of 20%. This technology is currently being developed at NASA Glenn Research Center.

The power management distribution system consists of MPCV-based boards housed in a rugged, space qualified VME enclosure. Cabling is based on Altair cable lengths but are resized for a 120 V bus and a 2% line losses. All power elements (masses shown in Table 6) are placed on the second stage.

**Table 6. Power Subsystem Masses.**

Item	Qty	Unit Mass (kg)	Basic Mass (kg)	Growth (%)	CBE Mass (kg)
<b>Dry Mass</b>					
<b>Power System</b>			<b>283</b>	<b>33</b>	<b>377</b>
Fuel Cell	3	41	123	25	154
Power Distribution Unit	2	32	64	25	80
Reactant Feed Lines	1	4	4	25	5
Cabling & Connectors	1	92	92	50	138
<b>Non-Propelled Fluids</b>					
<b>Fuel Cell Reactants</b>			<b>278</b>	<b>20</b>	<b>333</b>
O <sub>2</sub>			56	20	67
CH <sub>4</sub>			222	20	267

#### D. Thermal

The MAV Thermal Control Subsystem (TCS) performs three main functions: (1) mitigate heat loads and losses due to spacecraft interaction with the environment; (2) provide heat rejection for MAV subsystems such as Avionics, Power, ECLS, Human Factors (including crew metabolic heat), and Thermal Control; and (3) provide propellant conditioning for the MAV propulsion system during flight and surface storage. The TCS must operate in a wide range of thermal conditions, such as the diurnal Mars surface environment with its warm and cold extremes, the fairly benign Earth elliptical orbit, and the rather cold Mars orbit/transit environments.

The MAV TCS is composed of insulation systems, heaters and thermal coatings to moderate heating rates to and from the environment. Fluid systems, such as pumped coolant loops and cryocoolers with broad area cooling tubing networks, are included to perform heat collection. Loop heat pipe radiators are used to reject heat to the environment. Cryocooler and loop heat pipe technologies are currently not at a mature state of readiness. Fault tolerance for TCS elements is provided at the component level for such things as rotating equipment, valves and sensors.

MAV insulation systems, heaters, and thermal coatings are used to mitigate the impact of environmental heat loads on the crew cabin, the MPS propellant tanks and the vehicle as a whole. The crew cabin insulation is layered composite insulation (LCI) with a black Kapton outer layer. Shell heaters are used to maintain the internal cabin at room temperature. The propellant tanks are nested pairs of LCH<sub>4</sub>/LOX cryogenic storage tanks surrounded by vacuum jackets (VJs). The VJs have internal multi-layer insulation (MLI) and external LCI with an outer layer of Aluminized Teflon. LCI is installed in the space between the LCH<sub>4</sub> and LOX tanks. Broad area cooling tubes are installed under the MLI on the external tank walls with a high degree of thermal contact.

Pumped coolant loops (working fluid water/glycol) collect heat loads from MAV subsystems, located mainly within the crew cabin, and transport them to the external heat exchanger (i.e. evaporator) interface with the loop heat pipe radiator system. The loop heat pipe working fluid (NH<sub>3</sub>) is vaporized in the evaporator and flows to the onboard space radiator/condenser panels, where the working fluid is condensed and heat is rejected to the environment. Condensed working fluid flows back to the evaporator.

Cryocoolers (working fluid Ne) are multi-component systems which intercept heat loads penetrating the insulation system surrounding the propellant tanks via the broad area cooling tubing networks on the tank walls. Heat is rejected to the loop heat pipe evaporator, as with the pumped coolant loops. Based on operational constraints, it is assumed that each cryocooler will lift (remove) 100 W at 90 K. Active cryocoolers will maintain redundant components/units in a standby condition until needed.

The MAV TCS is fully integrated with the Mars Descent Module (MDM) TCS, and the two operate as one subsystem throughout the outbound flight and Mars surface stay until Mars launch and ascent (when the MAV flies independently). The MAV benefits from the afforded cooling and heat rejection load-sharing across the wide range of thermal environments, especially in the Mars surface environment. This approach also allows for the MAV to carry a minimum amount of TCS subsystem hardware, with the majority being left behind on the MDM at Mars launch. Table 7 gives available MAV and MDM cryocooler lift and active radiator area. Table 8 lists the thermal system masses required for heat rejection. These are grouped on the second stage and do not include the propulsion-related thermal elements that are given above in Tables 3 and 4.

**Table 7. MAV and MDM Heat Loads, Cryocooler Lift and Active Radiator Area.**

Mission phase	Available cryocooler lift, MAV (W)	Available cryocooler lift, MDM (W)	Available radiator area, MAV (m <sup>2</sup> )	Available radiator area, MDM (m <sup>2</sup> )
Outbound flight	100	100 (300 standby)	32	29 (stowed)
Mars surface stay	100	400	32	130 (deployed)
Mars orbit, MAV only	100	-	32	-

**Table 8. Thermal Subsystem Masses.**

Item	Qty	Unit Mass (kg)	Basic Mass (kg)	Growth (%)	CBE Mass (kg)
<b>Dry Mass</b>					
<b>Thermal Control</b>			<b>434</b>	<b>25</b>	<b>542</b>
Active Cooling Loops	1	180	180	25	225
Cabin Shell Heaters	1	10	10	25	13
Radiators/Sublimators	1	108	108	25	134
Cabin External Insulation	1	136	136	25	170
<b>Non-Propelled Fluids</b>					
<b>Coolants, Active Cooling Loops</b>			<b>50</b>	<b>25</b>	<b>63</b>

During the outbound flight and Mars surface stay, the combined MAV and MDM TCS heat collection/rejection capacity is sufficient to dissipate all MAV environmental and subsystem heat loads. At the time of Mars liftoff, the propellant tanks receive the high heat loads of the Mars surface environment when the cooling and heat rejection resources of the MDM TCS are suddenly unavailable. Due to the expected heat soakback from the tanks' insulation systems, propellant subcooling prior to Mars launch may be required to maintain appropriate propellant conditions after Mars launch.

#### E. Avionics

Avionics for the MAV are based on the Altair lunar lander ascent module avionics systems<sup>9</sup>. The Altair lunar ascent module had a very similar mission profile: ascend with 4 crew members and dock to an orbiting spacecraft. Key design components are a distributed command and data handling system using common avionics boxes, guidance, navigation and control systems to enable rendezvous and low impact docking with the orbiting spacecraft. Communications and tracking systems are also included. Table 9 summarizes the masses assumed for the avionics system. Future work will involve designing an avionics system specific to the MAV.

**Table 9. Avionics Masses.**

Item	Qty	Unit Mass (kg)	Basic Mass (kg)	Growth (%)	CBE Mass (kg)
<b>Avionics</b>			<b>340</b>	<b>20</b>	<b>407</b>
Command and Data Handling	1	238	238	20	286
Communication and Tracking	1	46	46	25	57
Guidance, Navigation, and Control	1	56	56	15	64

#### F. ECLS

Like the avionics system, the life support systems for the Mars Ascent Vehicle are based heavily on the Altair life support systems<sup>10</sup>. Some changes are made to the water system to include additional capability for purification due to the long duration of dormancy prior to crew access on the Martian surface. Additionally, nitrogen and oxygen supplies for cabin atmosphere are increased to allow for a small degree of cabin leakage during the vehicle's long duration of dormancy. The ECLS masses are shown in Table 10.

**Table 10. ECLS Masses.**

Item	Basic Mass (kg)	Growth (%)	CBE Mass (kg)
<b>Dry Mass</b>	<b>236</b>	<b>26</b>	<b>298</b>
<b>Non-propelled Fluids</b>	<b>196</b>	<b>0</b>	<b>196</b>
Nitrogen	21	0	21
Oxygen	17	0	17
Water (LCG)	13	0	13
Water (Sublimator)	109	0	109
Water (Potable)	36	0	36

## G. Human Factors/Cargo

This section describes the MAV payload (or cargo), which includes the crew, their support equipment and consumables (other than ECLS items), and the returned science samples. The type and quantity of Human Factors equipment needed is a function of a vehicle's crewed duration. Unlike longer-duration vehicles, the MAV's relatively short 2-day operational life allows the omission of many standard crew comfort items, such as a food warmer, potty, and exercise equipment. MAV Human Factors mass is best characterized as being limited to consumables and safety gear.

The MAV consumables include food, hygiene supplies (such as wet-wipes), and crew-worn items such as Maximum Absorbency Garments (MAGs). Potable water and breathing gasses are assumed to be part of the ECLS non-propellant fluids. Food consumption is based on a 1.831 kg per crew member per day requirement, including food wrappers plus a stowage bag to secure the food.

Safety gear includes personal radiation dosimeters, cabin illumination, a tool kit for contingency operations (such as a jammed hatch mechanism), a clean-up kit, and recumbent seating. For the purpose of this exercise, recumbent seats are assumed to be similar to the Orion project's seats and are by far the single largest Human Factors allocation at 22.7 kg each. Although MAV ascent acceleration loads are considered relatively gentle for a healthy crew launching from Earth, recumbent seating protects for two contingency scenarios: early return of deconditioned crew, or an incapacitated crew member. Forward work on these contingencies may offer mass reduction opportunities. All crew and cargo mass assumptions are listed in Table 11.

**Table 11. Cargo Assumptions.**

Item (# items)	Assumption
Crew (4)	98.5 kg/person
Food + Baggage	1.831 kg/person/day + 1.56 kg
Daily Crew Provisions	0.8825 kg/person/day
Total Mission Provisions	25 kg/person + 87.2 kg
Crew Transfer Bags	30.4 kg
Safety	39 kg
Recumbent Seats (4)	22.7 kg/seat
Umbilical Interfaces (2)	10.5 kg/interface
11 ft Umbilicals (4)	9.07 kg/umbilical
IVA LEA Suit (4)	15.5 kg/suit
Total MAGs	2.6 kg
Sample Container (10)	1.1 kg/container
Samples	239 kg

*Total Mass: 1,106 kg*

## VI. Design Sensitivities and Variants

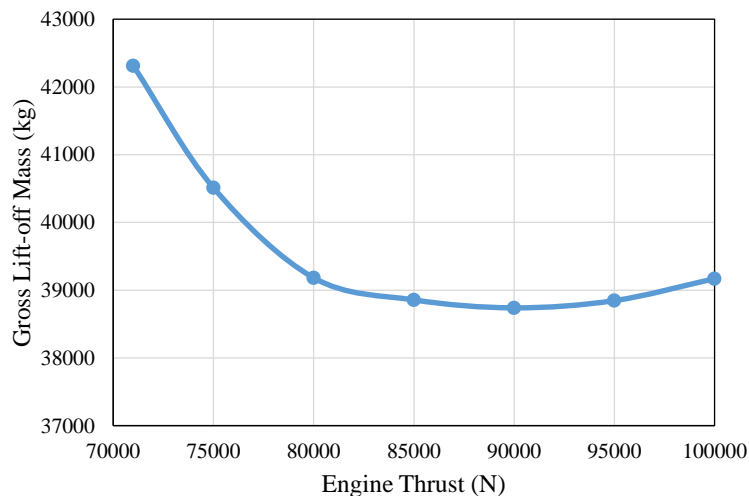
### A. Sensitivities to Ascent Flight Time

Crew consumables (food, water, and oxygen) mass and volume obviously increase as ascent flight time increases, but two additional sensitivities to ascent flight time are noted within Human Factors. Beyond two days it becomes difficult to keep crew inside their space suits. At this point, a crew waste and hygiene compartment (potty) begins to trade favorably for volume and health risk rather than accumulating soiled MAGs in the crew cabin. However, a waste/hygiene compartment comes with a mass and power penalty, plus additional cabin "elbow room" to remove and stow the suits. Beyond eight days in microgravity, the crew will require exercise equipment to maintain physical conditioning, which drives cabin volume and mass for both the exercise equipment and the accessories (workout clothing, harnesses, special shoes, etc.) that may be required. If a larger crew cabin is needed for extended duration hygiene and exercise, there will obviously be structural mass penalties, as well as impact to virtually every other subsystem. For example, a larger cabin diameter will incur mass penalties for everything from thermal insulation to cover the extra cabin surface area, to longer cable runs between equipment.

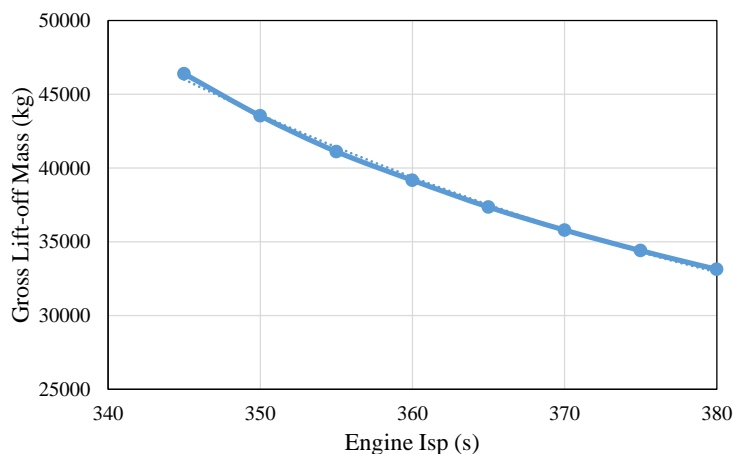
### B. Propulsion Sensitivities

The current propulsion system design is based heavily on trades performed in 2012 and documented in the Mars Design Reference Architecture 5.0 Addendum 2<sup>11</sup>. With the information gained in the latest design, new engine performance sensitivities are evaluated by optimizing trajectories using POST with the new mass information. Below are the results of two such trades. Figure 9 shows the effects of engine thrust on the MAV liftoff mass, while Fig. 10 shows the sensitivity to specific impulse. As expected, the mass decreases as the Isp increases. The thrust per

engine is slightly higher than optimal, but the higher thrust is tolerated so the same engine can be used on the descent stage, which benefits from the higher performance.



**Figure 9. MAV Liftoff Mass vs. Engine Thrust**



**Figure 10. MAV Liftoff Mass vs. Engine Specific Impulse**

### C. 5 sol MAV

There are two transportation architectures using SEP and chemical propulsion that are currently being evaluated as part of the Evolvable Mars Campaign. The first, the SEP-Chem Split, relies on purely SEP propulsion to deliver cargo to Mars and uses LOX/methane chemical propulsion stages to deliver crew to Mars and return them to Earth. For this option, the Mars parking orbit is the 1 sol case mentioned above. The second architecture, the SEP Hybrid, uses a combination of SEP and chemical propulsion on the same vehicle. These stages can deliver crew or cargo and may be reusable, returning to lunar vicinity for refueling when the primary mission is complete. The Mars parking orbit for this SEP Hybrid option is significantly larger at 250 km by 112,500 km altitude (orbital period of 5 Martian days, or 5 sols).

Using the 1 sol MAV as a reference, estimates were made on how the vehicle mass and functionality might have to change to reach the 5 sol orbit. The  $\Delta V$  needed to reach the higher orbit is the most obvious change (an additional 150 m/s) causing a growth in propellant mass and the associated tank volumes. Launch window availability and flight times are also affected. Because of the significantly larger parking orbit, launch windows may be available only once per week and flight times would typically range from 5-10 days. Options for shortening that flight duration were assessed. More analysis is needed but it appears there may be opportunities to launch and achieve rendezvous in 3 days with only 150 m/s more than the 1 sol case. These opportunities are likely to be very

infrequent. To achieve a 3-day flight time and preserve the option for daily launch windows, an additional  $\Delta V$  capability of nearly 500 m/s may be required.

The longer flight duration affects crew consumables and accommodations as discussed in section VI.A above. For the 5 sol MAV, it is assumed that a zero-g Waste Management System (WMS) or potty would need to be included at a mass of about 110 kg. In this design, crew members would still remain in their suits for the majority of the trip because the crew cabin is not large enough to accommodate crew plus suits, which when empty take up almost as much volume as another crew member. The MAV is not currently designed to accommodate a significant cabin atmosphere leakage event. With the crew members suited, they could quickly lower visors and safely continue ascent in an unpressurized cabin.

For the purpose of evaluating the SEP-Hybrid architecture, an assumption of a 3-day ascent and an addition of 150 m/s of propulsive capability was assumed. The total impact is 2.3mt growth in liftoff mass of the MAV and the loss of daily launch opportunities. Greater mass impacts may be realized with further study of rendezvous profiles.

#### **D. Crew Taxi (1 sol and 5 sol)**

Recent interest in exploration of Mars moon Phobos and Deimos<sup>12</sup> has sparked interest in using the MAV second stage (with integrated cabin) to transfer crew between the moons and the Earth return vehicle stationed in Mars orbit (either 1 sol or 5 sol orbit). Because this mission may require long loiter in Mars orbit and at least one year loiter near Phobos, the fuel cell power generation system is replaced with a solar array/battery combination, eliminating consumables. Replacing the power system results in a new inert mass of 7.2 t.

For a transfer from 1sol to Phobos and back, the required  $\Delta V$ s are approximately 2000 m/s for MPS and the 70 m/s for RCS and requires a propellant consumption of 6 t, which is well under the allowed maximum of 8.7 t. For the 5 sol mission, first the 1 sol MAV second stage propulsion system is scaled to cover the additional  $\Delta V$  to reach the higher orbit. Then, the power system is replaced (as mentioned above). Finally, the waste management system is upgraded to include the potty, and the crew consumables are increased to allow for a 3-day ascent. This results in an inert mass of 7.6 t. The total MPS and RCS  $\Delta V$ s are nearly identical to the 1 sol case, with different end point values. The resulting 5 sol taxi propellant requirement is 6.3 t. This is below the estimated maximum of 8.8 t. Recent structural analysis indicates additional changes would be required to support launching from Earth with full LOX propellant tanks since ISRU LOX production would not be available for this mission. These changes will be incorporated in future design iterations of the MAV-based Crew Taxi.

### **VII. Technologies Required**

One of the main reasons for doing studies of missions so far in the future is to identify necessary capabilities and technologies. This is used to inform technology investment prioritization so that mature technologies exist when vehicle development begins. A number of capabilities will be required to support a human Mars mission architecture. Table 12 identifies enabling technology needs for the Mars Ascent Vehicle. Additional development will be required for supporting mission elements, segments, and operations.



**Table 12. Required Technology Development.**

Topic Area	Category / Technology	Description
Avionics	Avionics software and Autonomous Vehicle Systems Management	<ul style="list-style-type: none"> <li>Management of onboard control, navigation, &amp; communication systems for rendezvous and proximity ops</li> <li>• Component-based Software Architecture (e.g., Core Flight System-CFS)</li> <li>• Wireless Solutions</li> <li>• Common Modular Hardware Building blocks (e.g., Space Frame)</li> <li>• Automated/Autonomous Rendezvous &amp; Docking</li> <li>• In-Space Timing and Navigation for Autonomy</li> <li>• Target Relative Navigation and Hazard Avoidance</li> </ul>
	C&T/C&DH	<ul style="list-style-type: none"> <li>Management of communication, data, and display systems</li> <li>• High Data Rate Forward Link (Flight) Communications</li> <li>• High Rate, Adaptive, Internetworked Proximity Communications</li> <li>• Delay Tolerant Network Software</li> <li>• Time Triggered Protocol (TTP/C)</li> <li>• Optical Communication</li> <li>• Optical Ranging</li> <li>• RF Imaging System</li> </ul>
ECLSS	Life Support	<ul style="list-style-type: none"> <li>High reliability, environmental control and life support systems for habitable volumes and EVA</li> <li>• High pressure oxygen</li> <li>• CO2 Removal and Advanced Environmental Monitoring</li> <li>• Universal Waste Management System<sup>†</sup></li> </ul>
Engine	LOx/LCH4 Cryogenic Propulsion	<ul style="list-style-type: none"> <li>Advanced development of integrated primary &amp; attitude control LOx/LCH4 Engine</li> <li>• Pump-fed LOx/LCH4 Propulsion (common engine development)</li> <li>• Integrated Pressure-fed LOx/LCH4 Reaction Control Systems</li> </ul>
EVA	EVA	<ul style="list-style-type: none"> <li>• Advanced space suit</li> <li>• EVA Tools &amp; Aids</li> <li>• Dust mitigation</li> </ul>
HRP	Biomedical Countermeasures	<ul style="list-style-type: none"> <li>• Risk mitigation for GCR and single event exposure</li> <li>• Long-term exposure to microgravity (support for deconditioned crew affects MAV design)</li> </ul>
ISRU	LOx In Situ Resource Utilization (ISRU)	<ul style="list-style-type: none"> <li>Enables a LOX-based economy resulting in a lower outbound vehicle mass and greater payload-carrying capability</li> <li>• Extraction of oxygen from CO2 in Martian atmosphere to provide propellant for Mars Ascent Vehicle (MAV) spacecraft on surface</li> <li>• NOT currently in baseline: Methane production at Mars supports long-term ISRU Strategy</li> </ul>
	Methane ISRU	
Power	Long-life Batteries*	• Low cycle, high specific energy for long duration energy storage
	Fuel Cells	• Low mass, high reliability, solid oxide, propellant scavenging of reactants
	Solar Arrays*	• Autonomously deployable w/ high strength/stiffness
	Fission Surface Power	• High reliability, continuous high power (kW) to support Lander(s) and ISRU reactor, power conversion, and power conditioning and distribution
	Structures and Mechanisms	• Lightweight, high-strength, materials, seals and mechanisms, and structural components
Thermal	Cryogenic Fluid Management and Storage / Thermal Control Systems	<ul style="list-style-type: none"> <li>Long-term storage and distribution technologies for cryogenics to enable zero boil-off (ZBO) for high performance cryogenic propulsion systems, surface systems (ISRU), and support power generation and</li> <li>• 90K Cryocooler</li> <li>• Multilayer Layer Insulation</li> <li>• In Space Cryogenic Liquid Acquisition</li> <li>• Micro gravity mass gaging</li> <li>• Thermal control for heat dissipation / heat rejection</li> <li>• Thermal Protection Systems</li> </ul>

\* Only required in MAV-based Crew Taxi variant

<sup>†</sup> Applicable to 5 Sol MAV and 5 Sol Taxi variants

## VIII. Conclusions

This work resulted in a significant improvement of our understanding of human Mars ascent vehicle requirements and design options as well as impacts of alternate missions like ascent to a 5 Sol orbit or serving as a crew taxi to and from Mars moons. Trajectory analysis led to a greater understanding of launch windows and performance sensitivities. Aerodynamic analysis resulted in a reduction of early drag estimates and a net savings of propellant mass. Propulsion system design and detailed propellant inventory improved upon earlier estimates and ensures that adequate propellant quantities are allocated. Detailed structures and power system designs replace previous parametric estimates with higher fidelity. Refined thermal analysis led to the addition of significant radiator area which affects RCS placement and placement of other lander payloads around the MAV. Propulsion

and thermal design efforts have also improved our understanding of required integration with the lander descent stage and ISRU plant. Evaluation of the effects of long duration dormancy led to improvements to ECLS systems, and crew equipment and cargo mass allocations have been reevaluated.

While tremendous progress has been made, we do not know enough yet to reduce the total lander mass from the original estimates which would reduce the challenging performance requirements for our Earth-to-Mars transportation vehicles. Additional work is needed to understand the requirements and mass of other payload elements that are delivered with the MAV such as the crew access tunnel and ISRU plant and to ensure feasibility of the current MAV design. A more thorough evaluation of MAV control systems is desired as well as an evaluation of dynamics of liftoff and staging and a preliminary design for stage separation mechanisms. A review of the avionics systems borrowed from the Altair lander design is needed as well. Continued efforts to refine the MAV design and the sensitivities to Mars architecture options will help to identify the most promising options for future human exploration and the necessary capabilities and technologies to make these missions possible.

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