

# **Modeling of Highly Instrumented Honeywell Turbofan Engine Tested with Ice Crystal Ingestion in the NASA Propulsion System Laboratory**

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The Propulsion Systems Laboratory (PSL), an altitude test facility at NASA Glenn Research Center, has been used to test a highly instrumented turbine engine at simulated altitude operating conditions. This is a continuation of the PSL testing that successfully duplicated the icing events that were experienced in a previous engine (serial LF01) during flight through ice crystal clouds, which was the first turbofan engine tested in PSL. This second model of the ALF502R-5A serial number LF11 is a highly instrumented version of the previous engine. The PSL facility provides a continuous cloud of ice crystals with controlled characteristics of size and concentration, which are ingested by the engine during operation at simulated altitudes. Several of the previous operating points tested in the LF01 engine were duplicated to confirm repeatability in LF11. The instrumentation included video cameras to visually illustrate the accretion of ice in the low pressure compressor (LPC) exit guide vane region in order to confirm the ice accretion, which was suspected during the testing of the LF01. Traditional instrumentation included static pressure taps in the low pressure compressor inner and outer flow path walls, as well as total pressure and temperature rakes in the low pressure compressor region. The test data was utilized to determine the losses and blockages due to accretion in the exit guide vane region of the LPC. Multiple data points were analyzed with the Honeywell Customer Deck. A full engine roll back point was modeled with the Numerical Propulsion System Simulation (NPSS) code. The mean line compressor flow analysis code with ice crystal modeling was utilized to estimate the parameters that indicate the risk of accretion, as well as to estimate the degree of blockage and losses caused by accretion during a full engine roll back point. The analysis provided additional validation of the icing risk parameters within the LPC, as well as the creation of models for estimating the rates of blockage growth and losses.

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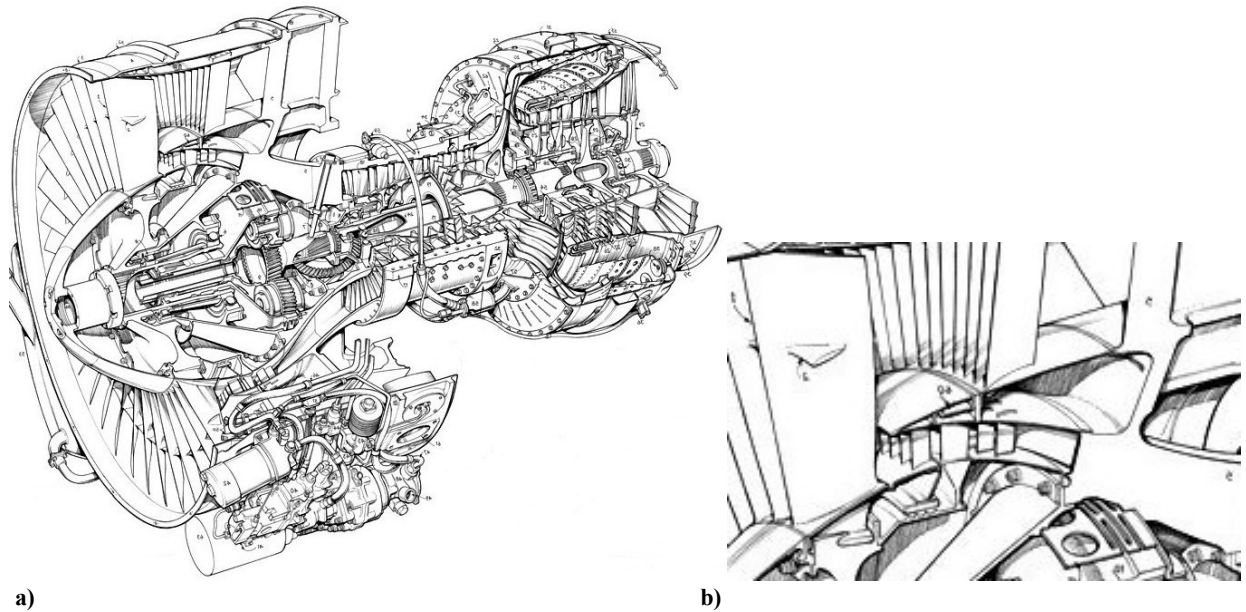
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## Introduction

THE highly instrumented turbine engine ALF502R-5A serial LF11 (Figure 1) has been tested in the Propulsion Systems Laboratory (PSL), an altitude test facility at NASA Glenn Research Center. The PSL facility provides a continuous cloud of ice crystals with controlled characteristics of size and concentration, which are ingested by the engine during operation at simulated altitudes. The purpose of the PSL testing is to improve understanding of the turbofan engine icing events<sup>1,2</sup> that have occurred in the turbofan engines of commercial airlines during flight through clouds with high ice water content. These events have been attributed to ice crystal ingestion and subsequent ice accretion in the compression system of the engine, which is responsible for the uncommanded loss of thrust (roll back). Testing of the LF11 engine is a continuation of the testing that was performed in 2013 on a similar engine (ALF502R-5A serial LF01)<sup>3,4,5</sup>.

This second model of the ALF502R-5A engine is a highly instrumented version of the previous engine. Several of the previous operating points tested in the LF01 engine were duplicated to confirm repeatability in the LF11. The instrumentation included video cameras to visually confirm the accretion of ice in the low pressure compressor (LPC) exit guide vane region, which was suspected during the testing of the LF01. Traditional instrumentation included wall static pressure taps as well as total pressure and temperature rakes in the low pressure compressor, exit guide vane and goose neck regions. Fifty-five Escort test data points were analyzed with the Honeywell Customer Deck (CD) and with a mean-line compressor flow code to estimate the values of key parameters that indicate the risk of accretion. Of those, twenty-two test data points were selected to highlight “threshold points” where the engine reached a partially degraded yet stable operating state, and the fastest roll back points which indicated that a roll back would be imminent (Appendix A). In addition, one full engine roll back point was modeled with the Numerical Propulsion System Simulation (NPSS) code to estimate the performance degradation of the LPC due to accretion. The NPSS<sup>6,7</sup> model of the LPC provided the boundary conditions for the mean-line compressor flow model (COMDES)<sup>8,9</sup> These two codes comprise the mixed fidelity computational tool that was utilized to provide details of the flow conditions within each blade row of the fan and low pressure compression system (LPC) during operation in the engine system environment.



**Figure 1: a) ALF502R-5A Turbofan Engine. b) The enlargement of the fan and single-stage LPC illustrating the IGV and the tandem stator; EGV stator 1 and EGV stator 2.**

Ice accretion in the LPC is a transient phenomenon and reduces the available aerodynamic area and deteriorates the performance of the compressor component where the accretion takes place, and consequently effects the overall performance of the engine. Modeling the full roll back data point at discreet Escort scan numbers with the mean-line compressor code required manually modifying the blockages and losses at the stator leading and trailing edges in the IGV and the tandem stator (EGV stator 1 and EGV stator 2). At each Escort scan number analyzed, the NPSS model results for pressure ratio and efficiency were utilized as boundary conditions for the compressor flow analysis. Additional losses in the IGV and EGV were required in the compressor as well as additional blockages were required

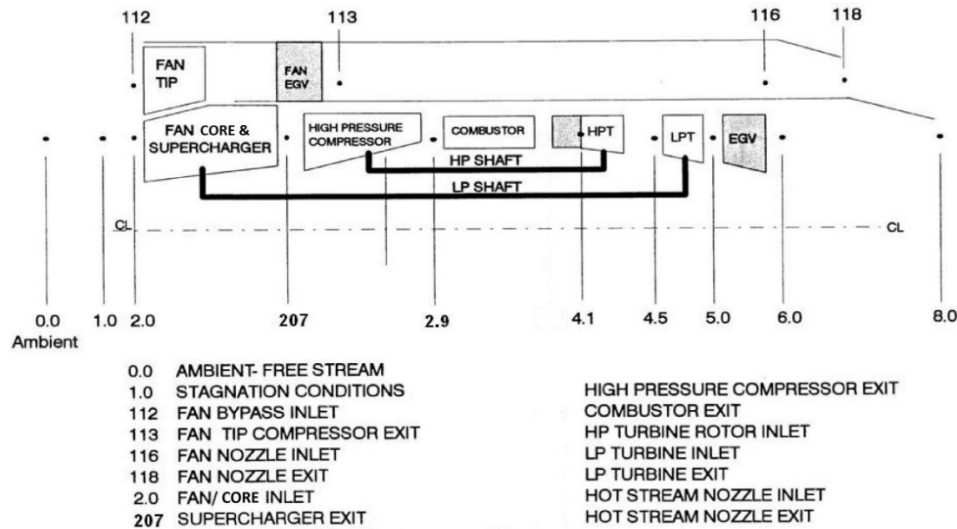
in the compressor flow model to match the measured static pressure at the flow path walls near the leading and trailing edges.

## I. Quantify Key Parameters Associated with Ice Accretion Risk

The first phase of this study involved the analysis of the select Escort data points in order to determine the values of key icing parameters associated with ice accretion, which for this small turbofan engine, could lead to engine roll back. The select test data points analyzed included both engine roll back and non roll back data points at a range of simulated altitudes. Note that for these roll back points the test engineer made the determination that the engine roll back would have been imminent. The imminent roll back was called in order to prevent possible damage due to ice shedding and impacting the high pressure compressor blades. For this reason, only one full engine roll back test point was conducted (Escort data point 093), which will be discussed in a subsequent section.

### A. Aerothermodynamic Simulation of the Engine System; Customer Deck and Compressor Flow Model

The customer deck (CD), that models the aerothermodynamic performance of the turbofan engine, was obtained from Honeywell Engines. The CD code provides the overall component-level performance and flow conditions, pressures, and temperatures, as illustrated in Figure 2. The fan is split into two distinct regions, the fan tip which models the flow through the bypass, and the fan core, which includes the low pressure compressor (supercharger). The CD system modeling code is comprised of component characteristic maps for each major engine component, and does not have the fidelity to provide the flow conditions within the stages and blade rows. Therefore a mean-line compressor flow code was utilized to provide the flow conditions at a higher fidelity within each blade row of the fan-core and low pressure compressor.

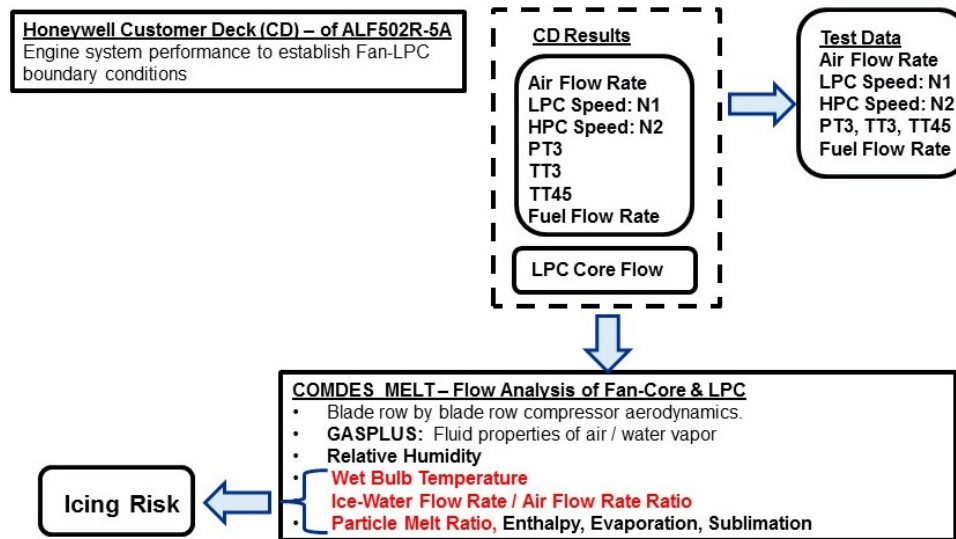


**Figure 2: The engine block diagram of the two spool turbofan engine, as modeled in the Customer Deck (CD).**

The compressor code includes a fluid properties model to account for supplemental water vapor in the air. Additionally, the computational tool incorporates a code named MELT<sup>9</sup> to model the melting of ice particles. The compressor analysis code computes the detailed blade row by blade row flow conditions, as well as the overall compressor performance. The MELT code has been developed to compute the state of ice crystals, as they pass through the engine inlet, fan and low pressure compressor blade rows. Models for sublimation, melting, and evaporation are included in the MELT code, however, there are no models for particle tracking, ice accretion, particle breakup, shedding, or erosion. The compressor code COMDES and MELT have been coupled to exchange boundary conditions at the blade leading and trailing edge of each blade row. The compressor code has a model to calculate the effects of water vapor on the fluid properties of the air-water vapor mixture based on the mole fraction of air to water vapor. The local relative humidity is computed through each component, as well as the sublimation, melting, and evaporation. The resulting performance of the compressor is computed based on the fluid properties of air with the effects of relative humidity.

The inputs provided to the CD system modeling code are the altitude, flight Mach number, fan physical rotational speed, and the air static temperature. The model results include the aerothermodynamic performance of each major engine component, as described in Figure 2, including the bypass ratio. The main purpose of the CD engine system model was to provide the bypass ratio and thus, the air mass flow into the engine core.

The COMDES and the CD system codes were utilized such that the boundary conditions from the CD at each operating point were passed as inlet boundary conditions to the compressor flow analysis code. A spreadsheet was created which facilitates a rapid means of providing the input into the CD code, code execution, passing the boundary conditions (BCs) to COMDES, and execution of it as well. The block diagram in Figure 3 illustrates the process by which the boundary condition data is transferred between the CD system model and the mean-line flow analysis code.

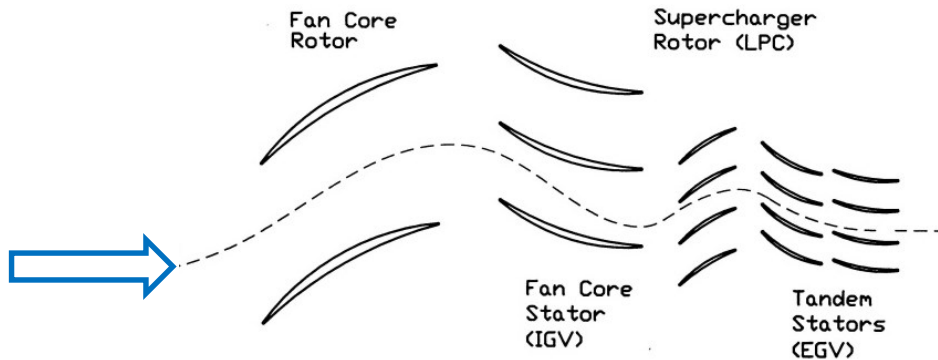


**Figure 3: Computational process using the Honeywell Customer Deck engine system model. The boundary conditions from the CD are passed to the compressor flow code to determine icing risk.**

The COMDES flow code computes the velocity, pressure, temperature, and flow angles at the leading edge and trailing edge of each blade row, at the hub, mean, and tip sections. The mean-line compressor flow analysis code has been modified to include the effects of relative humidity on the fluid properties of air and water vapor mixture, and the subsequent effects on compressor performance. The capability to calculate the local relative humidity in each blade row is based on the initial value of specific humidity (mass of water/mass of air) at the engine inlet, as well as the sublimation and evaporation of the particles through the flow path. In this engine, the location where ice accretes is suspected to be in the region of the exit guide vane (EGV) tandem stators (stator 1 and stator 2).

The ice particle melting and evaporation model, MELT, computes the local ice particle melt ratio and calculates the change of phase of an ice particle as it passes through the various components of the engine, starting at the engine inlet and through the fan and low pressure compressor blades and stator vanes. The ice particle residence times through each component are estimated in the MELT routine and are based on the velocities at the inlet and exit of the component and the distance the particle travels. The components are the engine inlet, each rotor and stator blade, support strut, and the axial gaps between the rotors, stators and the struts. The specific humidity, the ice water content (IWC), and particle diameter are specified at the inlet to the engine. Since there is no particle breakup model in the MELT code, the value for ice particle diameter in this study is assumed to be  $5\mu m$ , while the ice water content (IWC) is specified from the test data taken in the PSL. After the flow conditions through the blade rows have been calculated by COMDES, the static temperatures, pressures, and velocities are passed to MELT to determine the rate of melting, sublimation, and evaporation, and thus, the local ice particle melt ratio in each blade row. The calculations for sublimation, melting, and evaporation take into consideration the local static temperatures, pressures, and residence

times as they traverse through the engine inlet, the fan-core and low pressure compressor blade passages and gaps at the mid-span location, the gooseneck duct and the support strut. This requires an iterative exchange of boundary conditions between the COMDES and MELT codes as there is a change in fluid properties of the air as well as an exchange of enthalpy between the ice and the air.



**Figure 4: Ice particle path through the inlet, fan-core and low pressure compressor stage, including the tandem stator EGV (stator 1 and stator 2).**

Figure 4 shows a simplified illustration of the ice particle path through the rotor blades in the relative frame of reference, while the path through the stator vanes is in the absolute frame of reference. The calculations of ice sublimation, melt, and evaporation are performed from the leading to the trailing edge of each rotor and stator, as well as through the axial gap between the blades, as a function of velocity and distance traveled. The distances traveled are assumed to be equal to the chord length of each rotor and stator, as well as the gap between rotor and stator. The amount of water due to sublimation and evaporation is added to the local value of specific humidity, thus having an effect on the local relative humidity, and likewise on the local wet-bulb temperature calculation in the compressor code. The wet-bulb temperature calculation is based on the local value of static temperature.

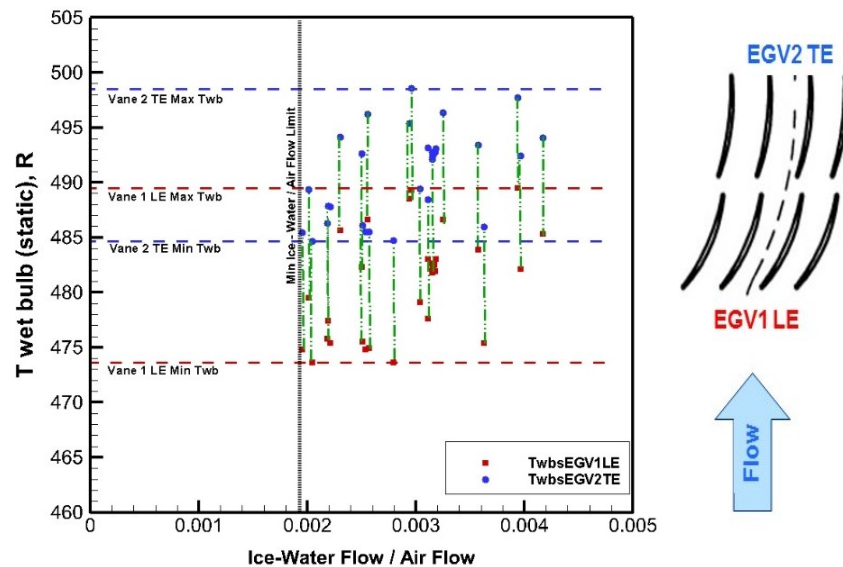
### B. Ice Accretion Risk Criteria

The targeted area in this engine where the initiation of ice accretion is suspected has been identified by the manufacturer as the tandem stator (EGV) region of the LPC, as illustrated in Figure 4. For the previous LF01 engine icing test in the PSL, the main focus of the computational analysis<sup>10,11</sup> was to validate the values of key parameters that indicate the potential for icing risk obtained from the NRC<sup>12,13</sup> data, and to further calibrate these parameters based on the subsequent analysis of data from the LF01 engine test. At that time only two parameters were considered: wet-bulb temperature and melt ratio. The tested engine times to roll back were categorized as: “fast” roll back, “slow” roll back, and no roll back, and the corresponding ranges of wet bulb temperature and melt ratio for each category were determined by analysis. Also it was assumed that there needed to be a minimum melt ratio (liquid to solid) so that the ice would accrete to the stationary parts of the engine components. Therefore the 5 $\mu$ m particle size was determined iteratively to “force” the analysis to result in a non-zero melt ratio in the EGV region when the engine rolled back. However, there were other sources of liquid water in the LF01 test present, mainly from the heated spinner and the anti-icing (heated) of the inlet guide vane (IGV). The additional sources of liquid water may have contributed to the ice accretion. The two key icing parameters which were identified as indicators of the risk of ice accretion were computed at each flow station. If the limiting values (determined from the LF01 test data) of these key parameters were met, there was a risk that ice would accrete on the surfaces of the compressor at that computing station. It was also observed in LF01 that there was a minimum value of ice-water flow rate to air flow rate ratio (*IWAR*), a dimensionless parameter, below which there was no risk of engine roll back.

$$IWAR = \frac{TWC * Q}{m}$$

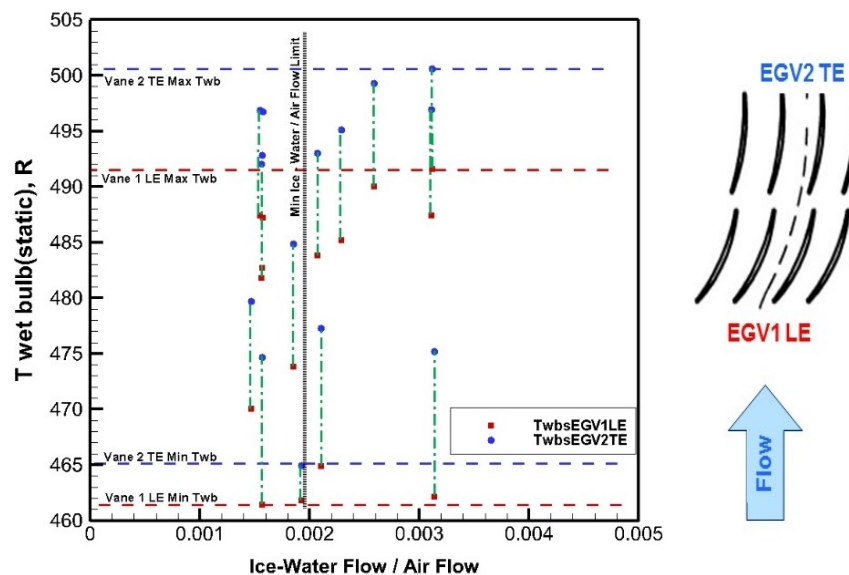
This minimum value was determined to be on the order of 0.002, or 0.2%. This minimum value was further confirmed in the current analysis of the LF11 test data.

Analysis of the test data from the highly instrumented LF11 engine enabled further calibration of the wet bulb temperature range between which there is a risk of ice accretion, potentially resulting in engine roll back. Figure 5 and 6 illustrate the calculated wet bulb temperatures for high altitude data points where the engine rolled back and also did not roll back.



**Figure 5: High altitude engine roll back points.**

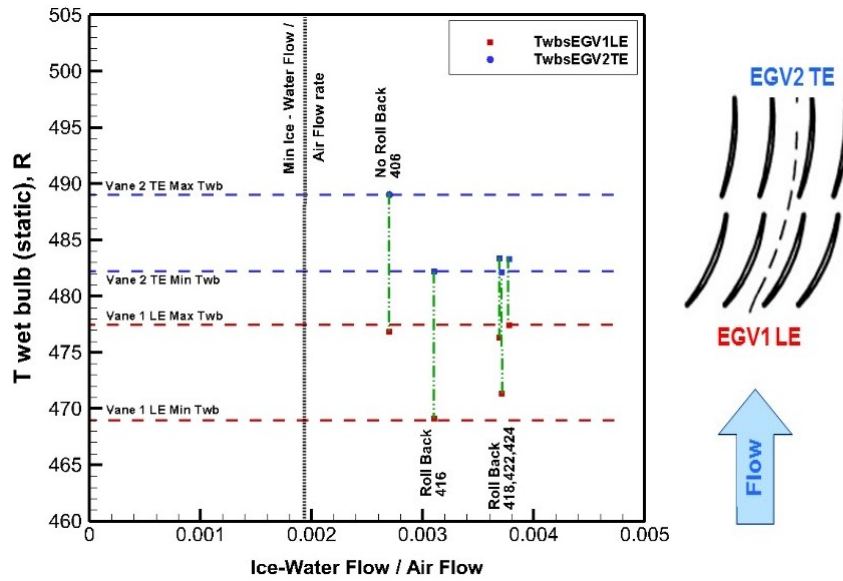
The values of wet bulb temperature at the EGV stator 2 trailing edge for the high altitude roll back points of Figure 5 appear to be clustered over the range 485 – 498 R. Also, the minimum value of the *IWAR* for engine roll back was approximately 0.002, and is similar to that noted in the previous LF01 engine testing.



**Figure 6: High altitude engine non roll back points.**

The values of wet bulb temperature at the EGV stator 2 trailing edge for the high altitude non roll back points of Figure 6 appear to be a wider range of 465 – 501 R. The lower value (465 R) is 20 degrees below the lower value of the points that rolled back (485 R), and the higher value (501 R) is 3 degrees above the higher value (498 R) of when the engine rolled back. Also, several of the points analyzed fall below the minimum value of the *IWAR* observed for the

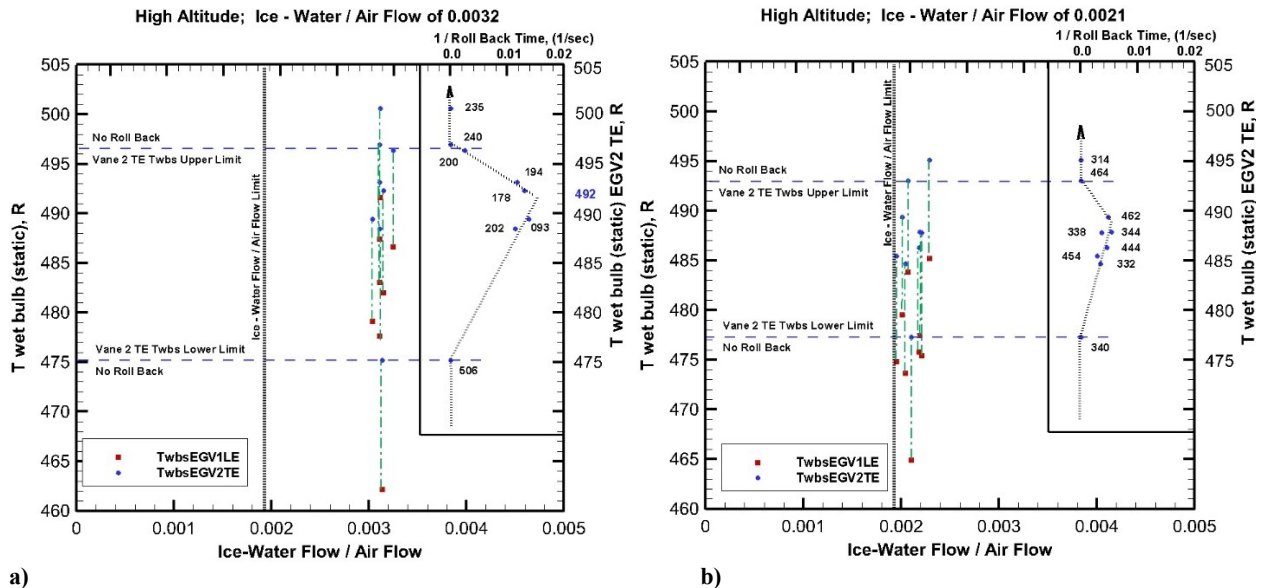
roll back points (0.002). Even though four out of six points were in the range of wet bulb temperatures that resulted in roll back, these were below the value of 0.002 of *IWAR*, thus there was not enough ice-water to cause the growth of significant amount of ice.



**Figure 7: Low altitude engine roll back and non roll back points.**

The LF11 engine was also tested at a simulated altitude of 4,800 feet to determine if it could be made to roll back in a similar way to the high altitude points. The inlet temperature was suppressed to a value significantly lower than the ISA temperature for this lower altitude. In addition, the flight Mach number, as well as fan speed were reduced. The conditions of the first point to be tested at 4,800 feet was such that the wet bulb temperatures at the stator 2 (Figure 7) fell within the range of the roll back points of Figure 5. However, the engine did not roll back, and there appeared to be slush and water running back on the stator 2 shroud. The inlet temperature was subsequently reduced, and the engine roll back was successfully accomplished. As a result, the calculated range of wet bulb temperature at the EGV stator 2 trailing edge is significantly lower (482 – 483 R) than at the higher altitudes. Escort data point 424 achieved a relatively fast called roll back in 138 seconds, thus demonstrating that the engine could be made to roll back in a similar time scale at low altitude, as at high altitude.

In an effort to understand the significance of the *IWAR* on the elapsed time for the engine to roll back, several data points were selected from Figures 5 and 6; in the vicinity of 0.0032 and 0.0021 and are plotted in Figures 8a) and 8b). In addition, the figures include a plot insert of the corresponding inverse of roll back time for each data point.

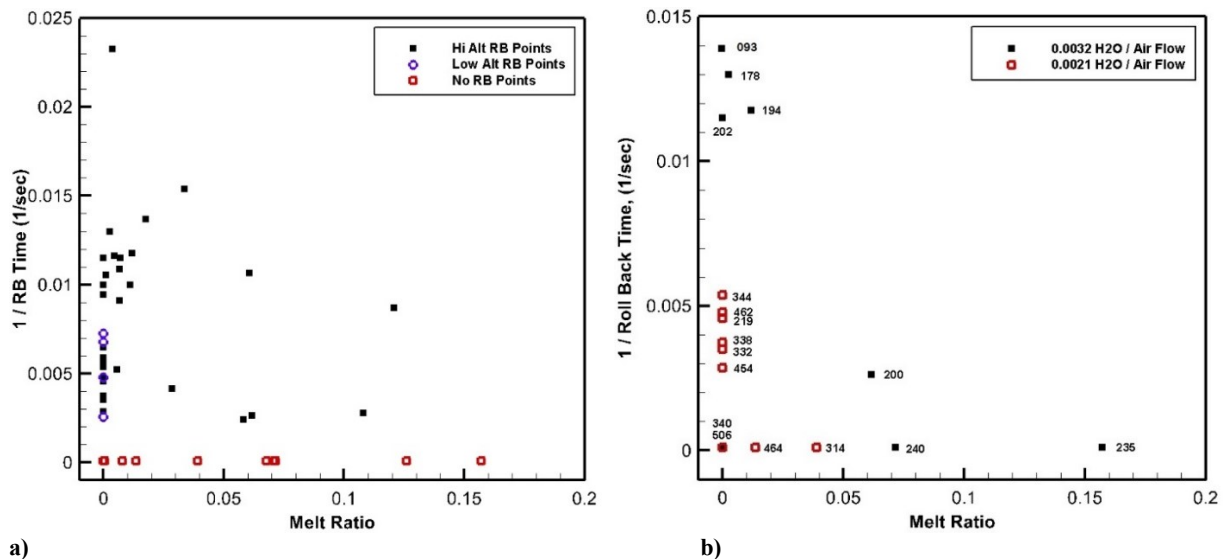


**Figure 8: High altitude data points at discrete ice-water flow to air flow rates (*IWAR*): a) 0.0032 b) 0.0021**

There appeared to be a strong correlation of wet bulb temperature and *IWAR* to the called roll back time. At an *IWAR* of 0.0032 (Figure 8a), the trend in roll back time appears to indicate that the fastest is at a wet bulb temperature near freezing (492 R). While at the lower value of *IWAR* (Figure 8b) the fastest roll back time appears to be at 489 R. At an *IWAR* of 0.0032 (Figure 8a), the upper threshold of wet bulb temperature for roll back is 497 R, and the lower threshold appears to be near 475 R. Likewise at an *IWAR* of 0.0021 (Figure 8b), the upper threshold of wet bulb temperature for roll back is 493 R, and the lower threshold appears to be near 477 R.

### C. Engine (Called) Roll Back Time vs. Ice Particle Melt Ratio

Figure 9 illustrates the relation between the calculated ice particle melt ratio, and the called engine roll back time. Note that the melt ratio is a calculated number based on an assumed particle size, and only considers heating from the air. There are other sources of liquid water that are not accounted for in the model, such as the heated spinner, and the melting due to kinetic energy of the particle hitting the compressor components. In addition the heat capacity of the warm metal surfaces of the compressor can also be a source of liquid water.



**Figure 9: Engine called roll back time (inverse), versus COMDES model estimate of particle melt ratio. a) All data points analyzed. b) Data points at *IWAR* of 0.0032 and an *IWAR* of 0.0021.**



The heated IGV is accounted for through the CD engine system model. As illustrated in Figure 9a), there are numerous data points that had a called engine roll back, even though they had zero particle melt ratio due to heating from the air. Likewise, there are data points that have a non-zero melt ratio, yet they did not experience an engine roll back. The reasons for this are that the wet bulb temperature was either too warm to support accretion, or there was not enough *IWAR*. Figure 9b) illustrates the called roll back time versus calculated particle melt ratio, for the two specific values of *IWAR* (0.0032 and 0.0021). These points are the same as those shown in Figures 8a) and 8b). Note that neither of the fastest roll back times at the two values of *IWAR* (Escort data points 093 and 344) show a non-zero calculated particle melt ratio at the EGV stator 2 trailing edge region. This may indicate that in order for the calculated melt ratio to be non-zero, the particle size may be even smaller than the assumed  $5\mu m$ . Concurrently, it may indicate that heating from the air does not significantly contribute to the liquid water required for accretion to take place. Alternatively, the other possible sources of liquid water (heated spinner, kinetic energy, heat capacity of the warm metal surfaces) may be responsible for ice accretion.

#### D. Engine Roll Back Time vs. Ice-Water Flow to Air Flow Rate (*IWAR*)

The significance of the *IWAR* on the elapsed time for the engine to roll back is further illustrated in Figures 10 a) in which the roll back time versus *IWAR* is plotted. The peak roll back times are highlighted as the upper limits of roll back times for *IWAR* of 0.0032 and 0.0021. The extension of the roll back time limit line beyond 0.005 is based on Escort data points 160 and 162, which had values of *IWAR* on the order of 0.007, and called roll back time of 34 seconds.

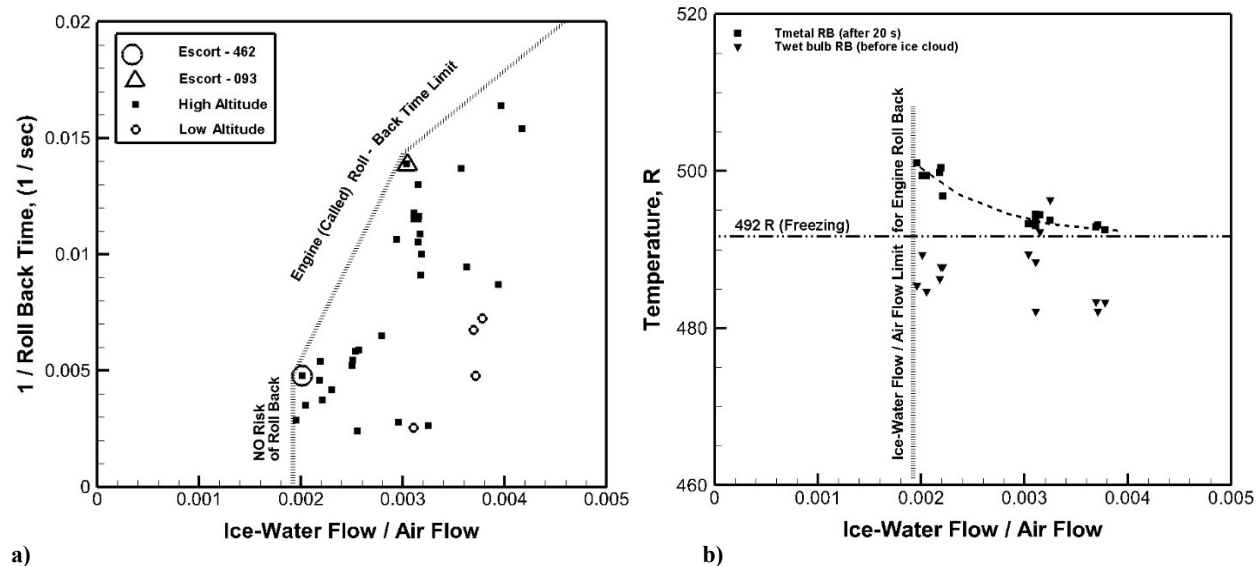


Figure 10: a) Engine roll back time versus *IWAR*. b) Measured EGV shroud metal temperature (after 20 seconds of ice cloud on) and calculated wet bulb temperatures (before ice cloud) versus *IWAR*.

Figure 10b) illustrates the shroud metal temperatures at EGV stator 2 trailing edge, 20 seconds after the ice cloud has been turned on, versus *IWAR*, at all the roll back points analyzed at all altitudes. It can be seen that there is a dependence of metal temperature on *IWAR*. A low value of *IWAR* below 0.002 cannot cool the metal significantly enough in order for ice accretion to occur. Concurrently, values of *IWAR* in the vicinity of 0.003 and greater can quickly cool the metal temperature to freezing. The calculated wet bulb temperatures at EGV stator 2 trailing edge prior to ice cloud on condition are also shown for reference in Figure 10b).

E. Images of Accretion at EGV stator 2 Trailing Edge at Ice-Water Flow to Air Flow Ratio (*IWAR*) of 0.0032

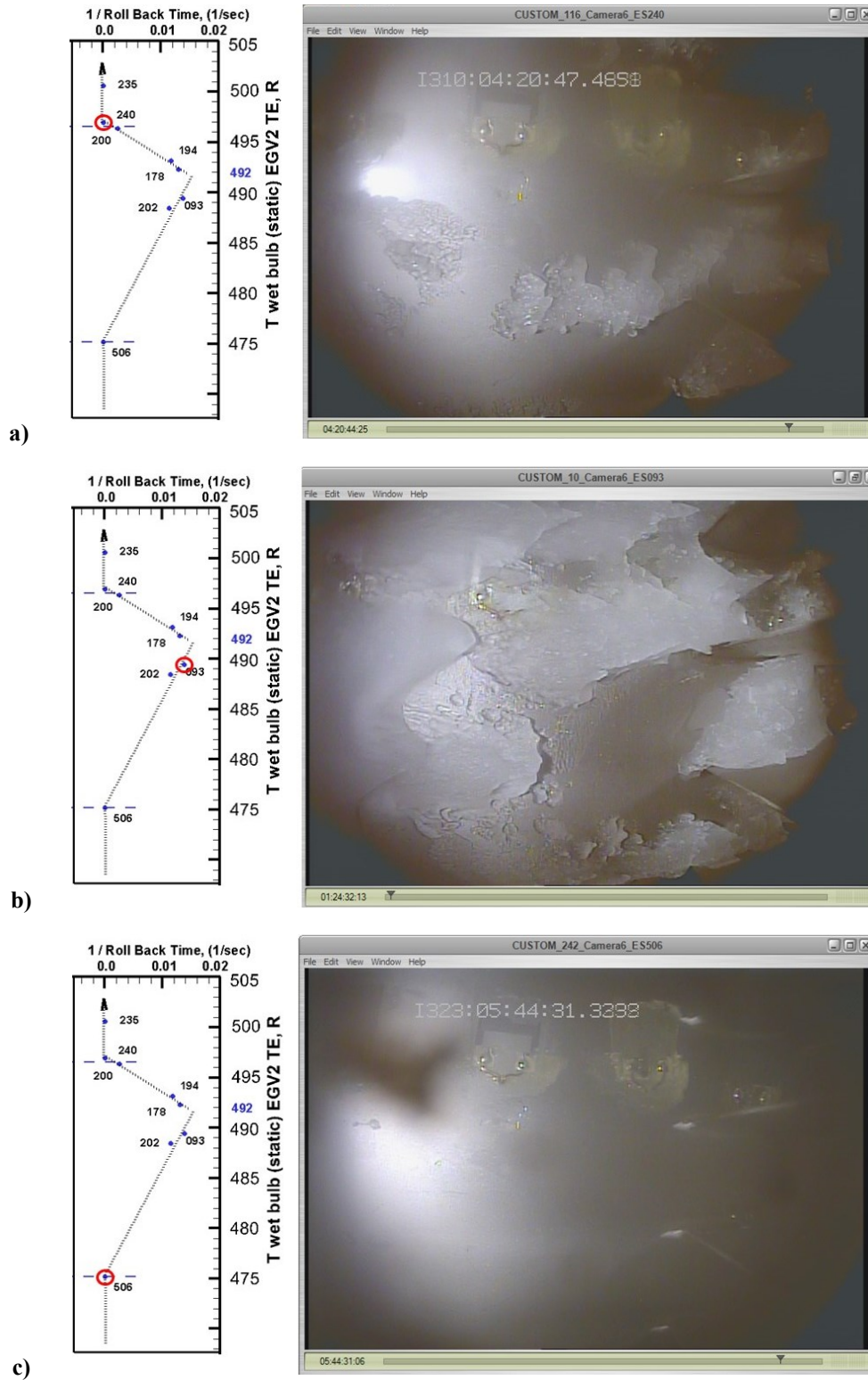
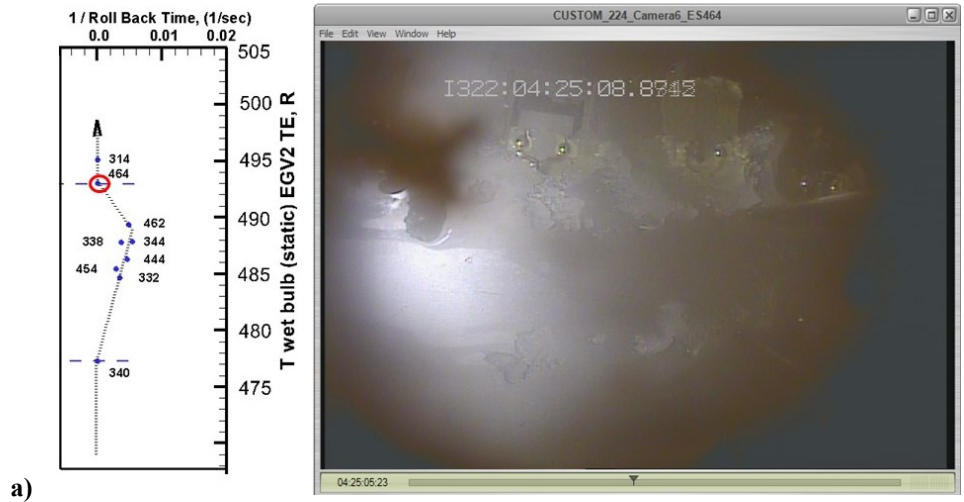


Figure 11: a) Max. Twet bulb threshold Point 240, b) “Max. Growth Rate Point” 093, and c) Min. Twet bulb threshold Point 506

The images shown in Figure 11 are of three Escort data points at *IWAR* of 0.0032. Figure 11a) is a no roll back data point (Escort point 240) that had a high calculated wet bulb temperature (497 R) at EGV stator 2 trailing edge. Even though the *IWAR* was acceptably high to support strong accretion in the blade passages, the ice was slushy and there was ice buildup and shedding taking place. The high value of wet bulb temperature is suspected to be the reason the ice did not grow at a significant rate over time, and consequently did not roll back. Figure 11b) represents the growth of ice for Escort data point 093 at Escort scan 72 (47 seconds after the ice cloud was turned on). Note that the wet bulb temperature at EGV stator 2 trailing edge was near freezing at this data point. There was strong ice accretion with minimal shedding, and the ice accretion continued to grow, resulting in significant blockage (near 60%) at Escort scan 305 (discussed later in Section II “Analysis of Full Roll Back Point”). Figure 11c) illustrates the EGV stator 2 trailing edge for Escort data point 506, showing that there was no ice accretion and no roll back occurred. The wet bulb temperature at this data point was too cold to support ice accretion, at 475 R.

**F. Images of Accretion at EGV stator 2 Trailing Edge at Ice-Water Flow to Air Flow Ratio of 0.0021**

The images shown in Figure 12 are of three Escort data points at *IWAR* of 0.0021. Figure 12a) is a no roll back data point (Escort point 464) that had a calculated wet bulb temperature (493 R) at EGV stator 2 trailing edge. The *IWAR* of 0.0021 is near the minimum limit to support accretion, and the small amount of ice that did accrete was slushy with shedding. The combined effects of relatively low value of *IWAR* and slightly elevated value of wet bulb temperature at the stator 2 trailing edge, are suspected to be the reasons the ice did not grow at a significant rate over time, and consequently did not roll back. Figure 12b) represents the growth of ice for Escort data point 462, which exhibited a slow called engine roll back (209 seconds). Note that the wet bulb temperature at EGV stator 2 trailing edge was slightly below freezing (489 R) at this data point. There was slow ice accretion with shedding, but the ice continued to grow until engine roll back was called by the test engineer. Figure 12c) illustrates the EGV stator 2 trailing edge for Escort data point 506, showing that there was small amount of ice accretion streaks which did not build up in time, and no roll back was called. The stator 2 trailing edge wet bulb temperature at this data point was too cold to support ice accretion, at 477 R.



a)

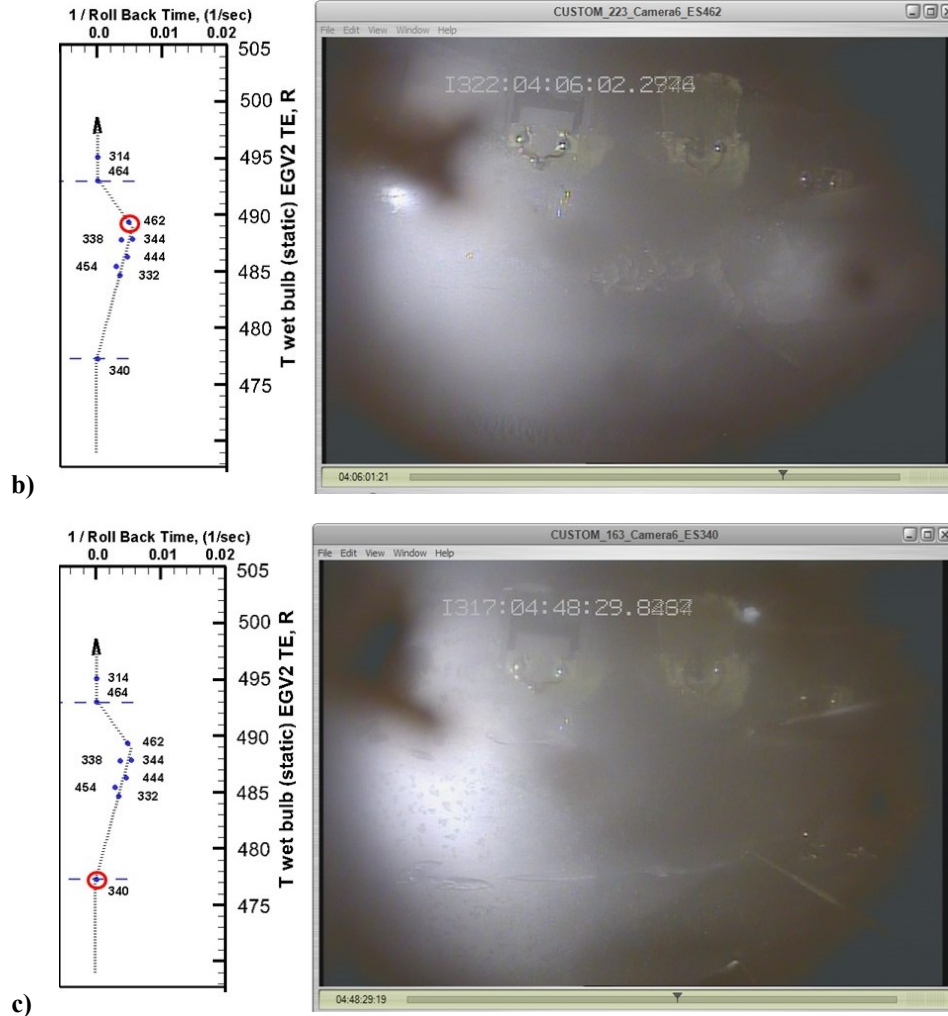
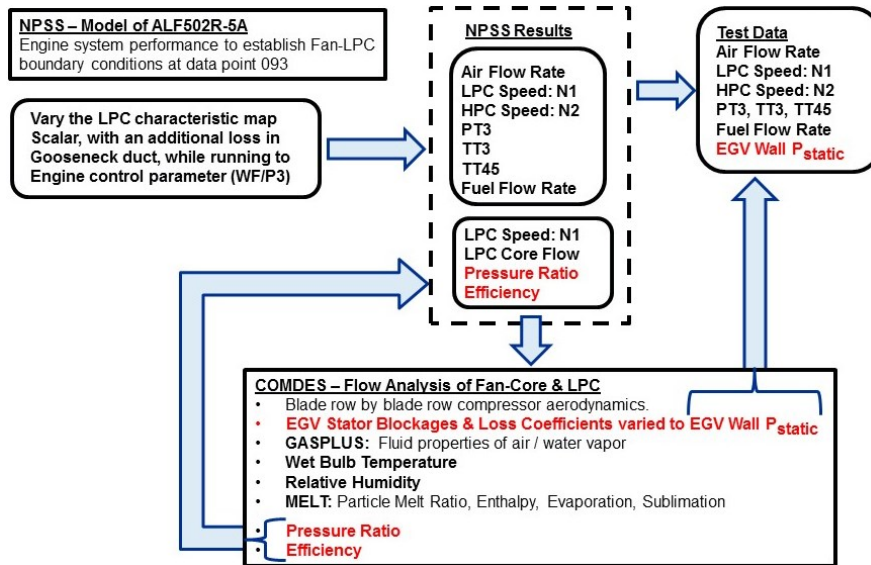


Figure 12: a) Max. Twet bulb threshold Point 464, b) “Max. Growth Rate Point” 462, and c) Min. Twet bulb threshold Point 340

## II. Analysis of the Full Roll Back Data Point (Escort 093)

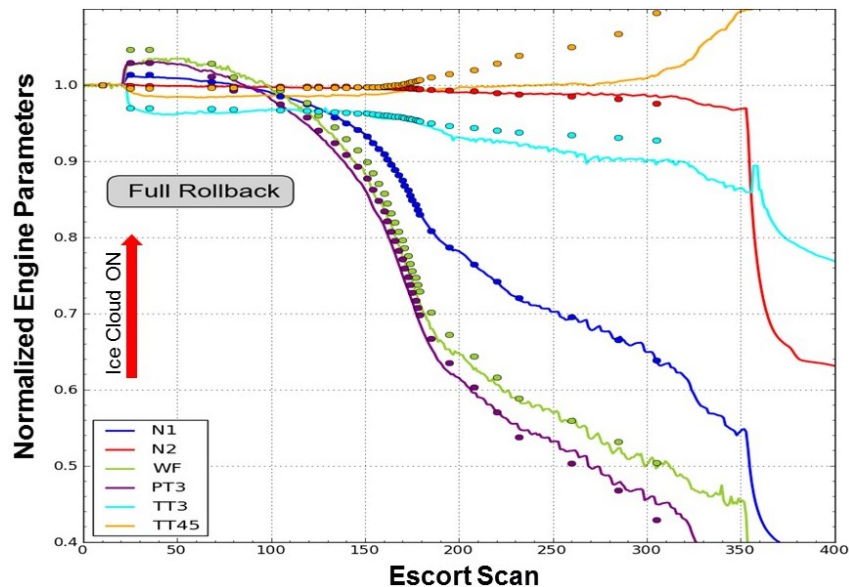
### A. Engine Performance Modeling with NPSS Compared to Measured Escort Scan Data

One full engine roll back point was modeled with the Numerical Propulsion System Simulation (NPSS) code to estimate the performance degradation of the LPC due to ice accretion. The objectives of this study were to determine the rate of change of the blockage growth rate, percent blocked area, pressure loss coefficients, and the incidence angles through the IGV, EGV stator 1 and EGV stator 2, through full engine roll back Escort data 093. The approach was to utilize the NPSS engine system modeling code such that the model matched the engine (Escort scan) test data, in order to determine the performance of the key engine components. The component characteristic maps of the fan-core and the LPC within the NPSS system model were manually degraded to match the tested engine performance, as a function of Escort scan numbers. The engine parameters that were matched as closely as possible are the fan speed (N1), core speed (N2), fuel flow rate (WF), core compressor exit pressure (PT3) and temperature (TT3), and the high pressure turbine exit temperature (TT45). The analysis procedure is similar to that utilized in the previous PSL engine icing test, namely the LF01 engine, and is outlined in more detail in an earlier publication<sup>13</sup> (indicated by dashed rectangle in Figure 13).



**Figure 13: Computational process using the NPSS system model with degradation of the LPC performance, and the compressor flow code with additional blockages and losses to iteratively match the test data results.**

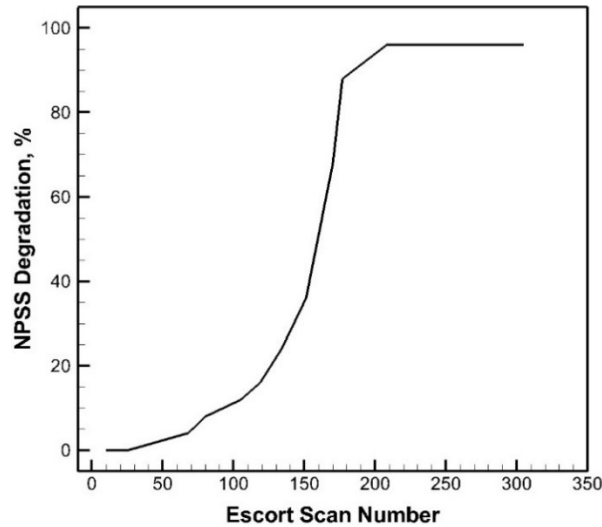
The results of this process of matching the engine parameters in the NPSS model to the measured Escort scan data are illustrated in Figure 14. The NPSS model results (symbols) closely match most of the measured engine parameters starting before the ice cloud is turned on (scan 10) and through the engine roll back up to Escort scan 305. Note that the ice cloud was turned on at Escort scan 25, where the engine immediately reacted to the added mass flow rate of the ice particles, and the NPSS model reflected the change without any modifications.



**Figure 14: NPSS performance analysis (symbols) of Escort data point 093 (full engine roll back), compared to test data (solid curves).**

In order to achieve the reasonably good match between the NPSS model results and the Escort test data after scan 30, a degradation of the fan-core and LPC pressure ratio and efficiency characteristics were required. The performance degradations were applied in 4% increments, up to 96% in order to match the measured engine performance data, and

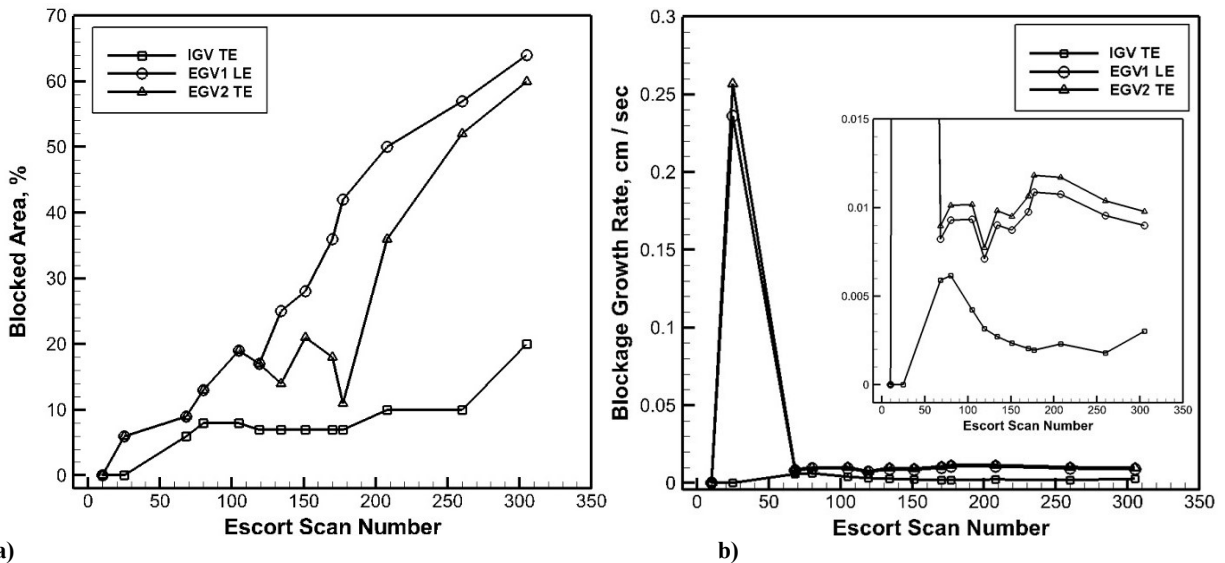
are illustrated in Figure 15. Note the applied degradations increased rapidly from 0% up to 96% and leveled out beyond scan 180.



**Figure 15: Degradation in NPSS of the fan-core and LPC pressure ratio and efficiency. This degradation was necessary in order to match the Escort test data throughout the full roll back.**

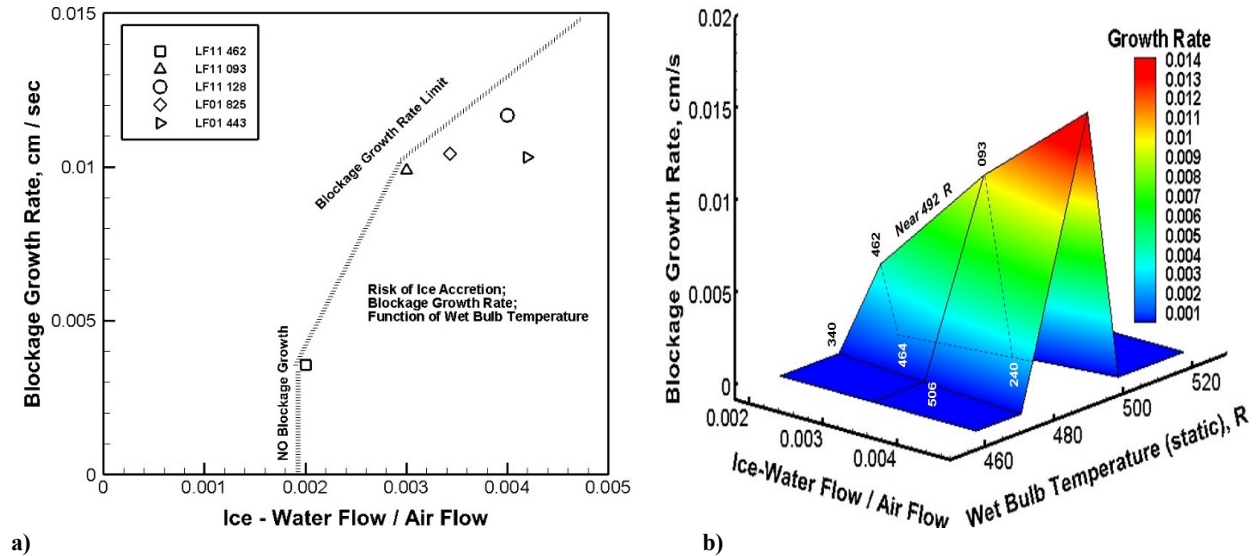
### B. Fan-Core and LPC Modeling with COMDES

In this section the mean-line compressor flow analysis code was subsequently used to determine the values of blockages and pressure losses in the IGV and EGV, that were required to match the overall pressure ratio and efficiency boundary conditions provided by the NPSS model. As part of this process, illustrated in Figure 13, the wall static pressure in the IGV and EGV from the Escort scan test data were also used as boundary conditions in the mean-line compressor flow model. The boundary conditions for the COMDES flow modeling process are highlighted in red font in Figure 13. The objective of modeling with the COMDES mean-line flow code was to determine the blockage growth rate due to accretion and boundary layer growth, and the corresponding pressure loss coefficient parameters. This process was applied at several discrete time steps through the full engine roll back data point 093 at Escort scan numbers 10, 25, 68, 80, 105, 119, 134, 151, 170, 177, 208, 260, and 305. Note that each scan is approximately one second.



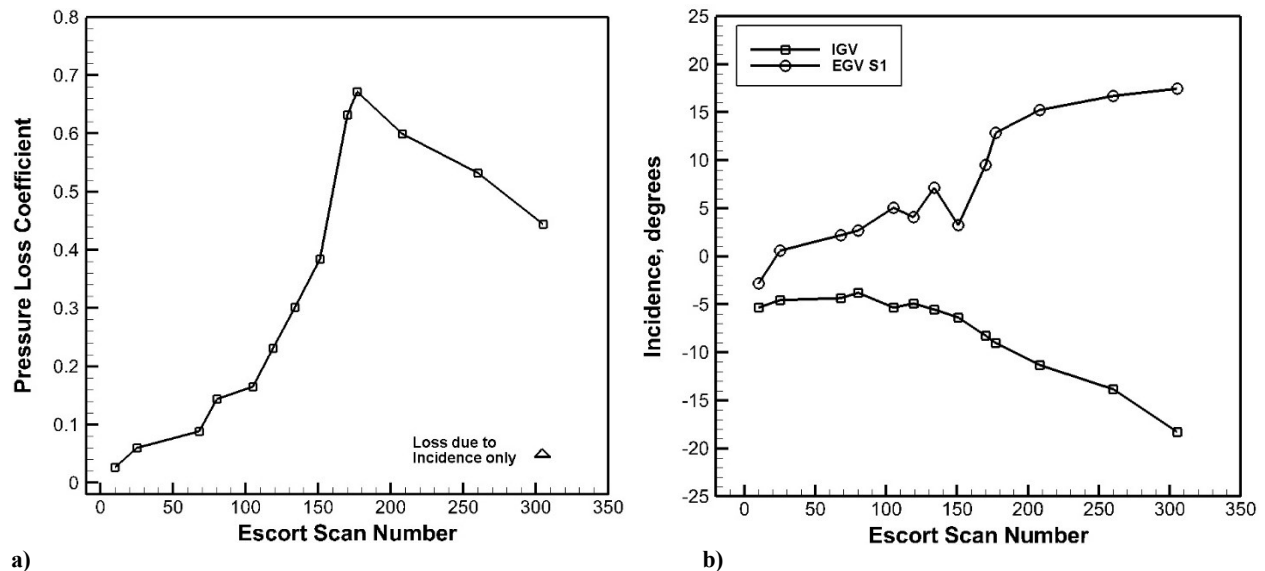
**Figure 16: Escort data point 093: a) Percent of blocked area versus Escort scan number. b) The moving average of blockage growth rate versus scan number. The insert is a zoomed view.**

The additional blockages for the IGV and the EGV that were necessary to match the LPC boundary conditions from NPSS and the Escort data are shown in Figure 16a). Note that the blockage at scan number 305 is on the order of 60% of the annular area. The moving average of the growth rate of blockage versus Escort scan number is illustrated in Figure 16b). At the end of scan number 305, the average blockage growth rate at the EGV stator 1 leading edge and stator 2 trailing edge is on the order of 0.01 cm/sec. This rate of blockage growth determined from the analysis of Escort data point 093 was then used as the basis for scaling the maximum limit of blockage growth rate at two other values of *IWAR*; 0.002 and 0.004, and is illustrated in Figure 17a). The basis for scaling the maximum blockage growth rate limit to the other two values of *IWAR* was the calculated growth rate for Escort data point 093.



a) Figure 17: a) Growth rate of blockage versus *IWAR*. b) Surface plot of growth rate of blockage versus *IWAR* and wet bulb temperature.

Figure 17b) shows the surface plot of growth rate versus wet bulb temperature and *IWAR* that was generated using the estimated values of blockage growth rate versus *IWAR* shown in Figure 17a), and the minimum and maximum threshold values of wet bulb temperature shown in Figures 8a) and b). The Escort data numbers are shown for each point defining the surface plot of Figure 17b).



a) Figure 18: a) Overall “total pressure loss coefficient” through the EGV stator 1 and EGV stator 2. b) Leading edge incidence versus Escort scan number.

Figure 18a) shows the additional blockage that was required in the compressor flow code to match the NPSS boundary conditions of the fan-core and LPC total to total pressure ratio and efficiency, as well as the wall static pressure measurements in the Escort data, as a function of Escort scan number (time). Note that the pressure loss coefficient of Figure 18a) tracks the performance degradation that was applied to the NPSS model (Figure 15). The total pressure loss coefficient is primarily due to the time-dependent deterioration of surface roughness caused by the ice accretion. The losses grew to a value of 0.66 near scan 180, then tapers off to 0.45 at the end of scan 305. This inflection point is caused by the drastic reduction of the air mass flow rate into the core. However, the local Mach number at the stator 1 leading edge and stator 2 trailing edge increases due to the large area reduction that was required to match the measured wall static pressures, even though the core mass flow rate decreases. The Figure 18b) shows the incidence versus Escort scan number, as estimated from the compressor flow analysis. Even though the EGV stator 1 incidence is high at Escort scan 305, the component of pressure loss coefficient due to incidence alone is only 0.05 (See Figure 18a).

The fan-core performance was modeled with the compressor flow code prior to the ice cloud initiation, and the performance results compared well to the Escort measurements of total pressure and temperature ratios, as well as the wall static pressures of EGV stator 1 and stator 2. However, once the ice cloud was turned on, the total pressure and temperature measurement instrumentation were ineffective due to fouling by the water and ice. Therefore, once the ice cloud was turned on, the compressor model results were utilized to estimate the air total pressure and temperature through the engine roll back. Illustrated in Figure 19 are the model results for the total and static temperatures, as calculated by the compressor model for the shaft speed and core mass flow rate obtained from the NPSS system model. Also included in Figure 19 are the calculated values of wet bulb temperature and the measured shroud metal temperatures at the EGV stator 2 trailing edge region. From this it can be seen that through the engine roll back, all the temperatures change significantly from their initial values prior to ice cloud on, yet the growth of blockage (partially due to ice accretion) continues unimpeded through Escort scan 305, as was observed in Figure 16a).

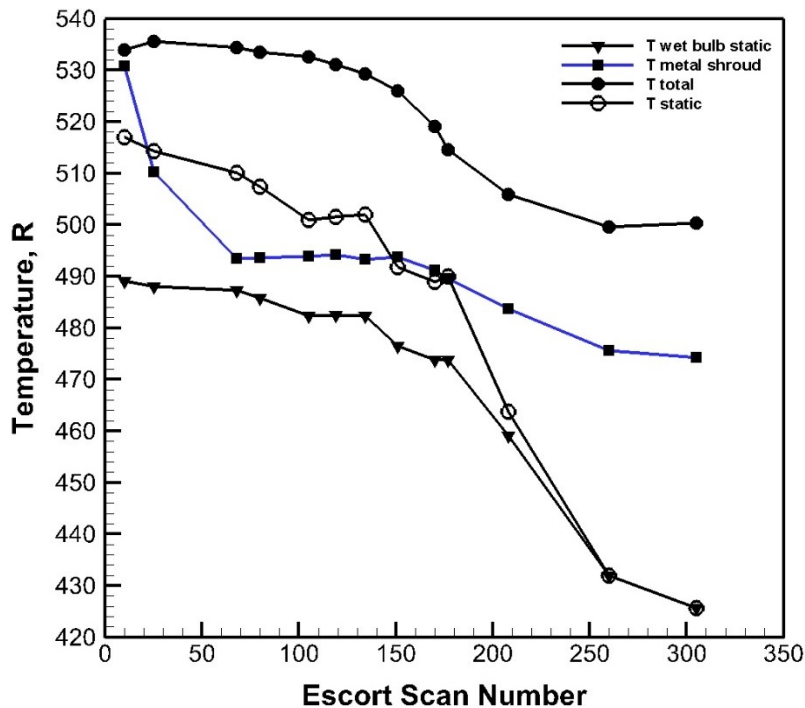
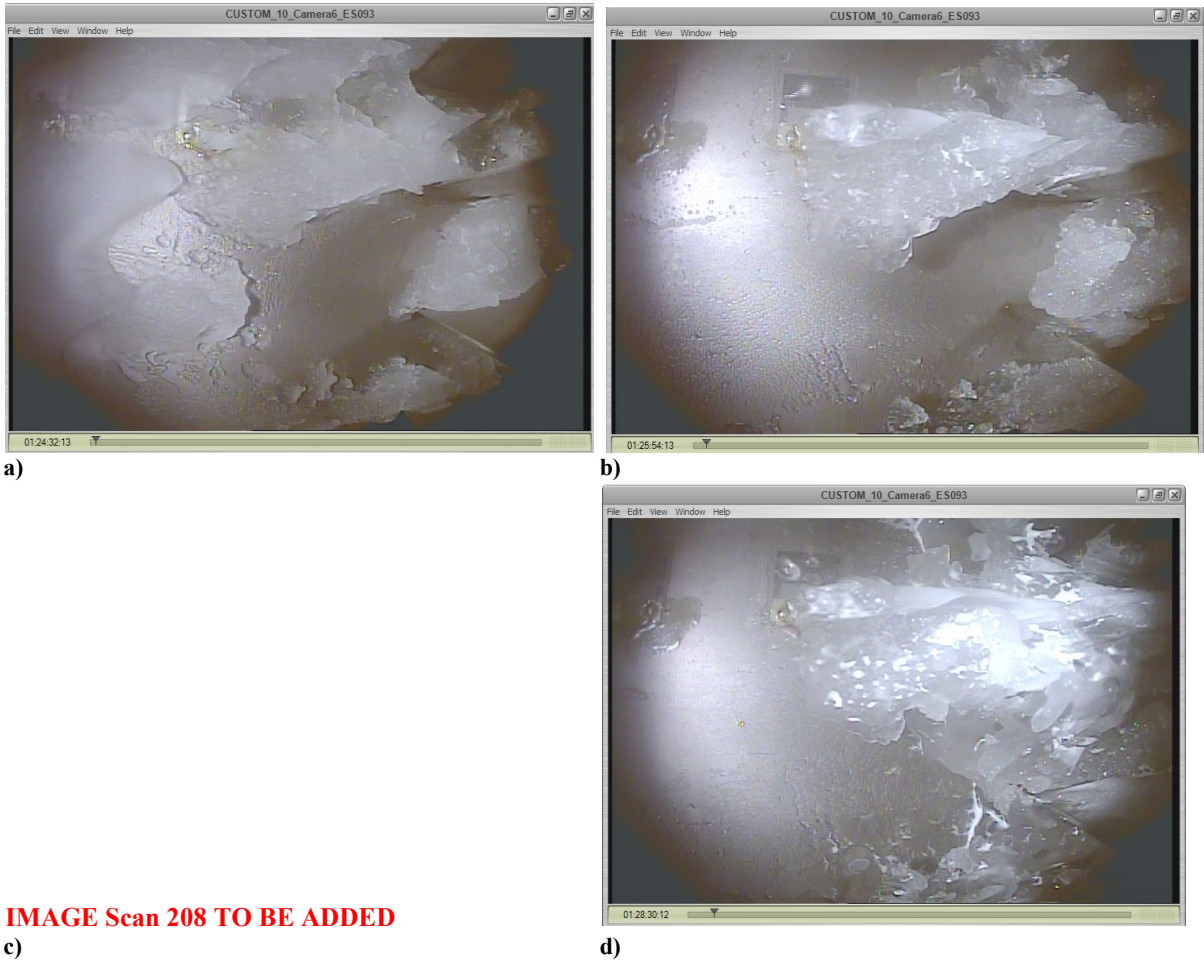


Figure 19. Escort data point 093. EGV stator 2 temperatures: wet bulb, shroud metal, air total, air static; versus Escort scan number.



### C. Images at EGV stator 2 trailing edge during full engine roll back (Escort data point 093)

Escort data point 093 was a full roll back point that was tested with ice cloud on for over 300 seconds, but it would have met the criteria for a “called roll back” at Escort scan 72, which was 47 seconds after the ice cloud was turned on. This was done in order to compare its performance to the other called roll back points. Figure 20 further expands on Figure 11b) to illustrate the ice growth at Escort scans 150 and 305. (Figure 20b) is the same as Figure 11b).



**IMAGE Scan 208 TO BE ADDED**

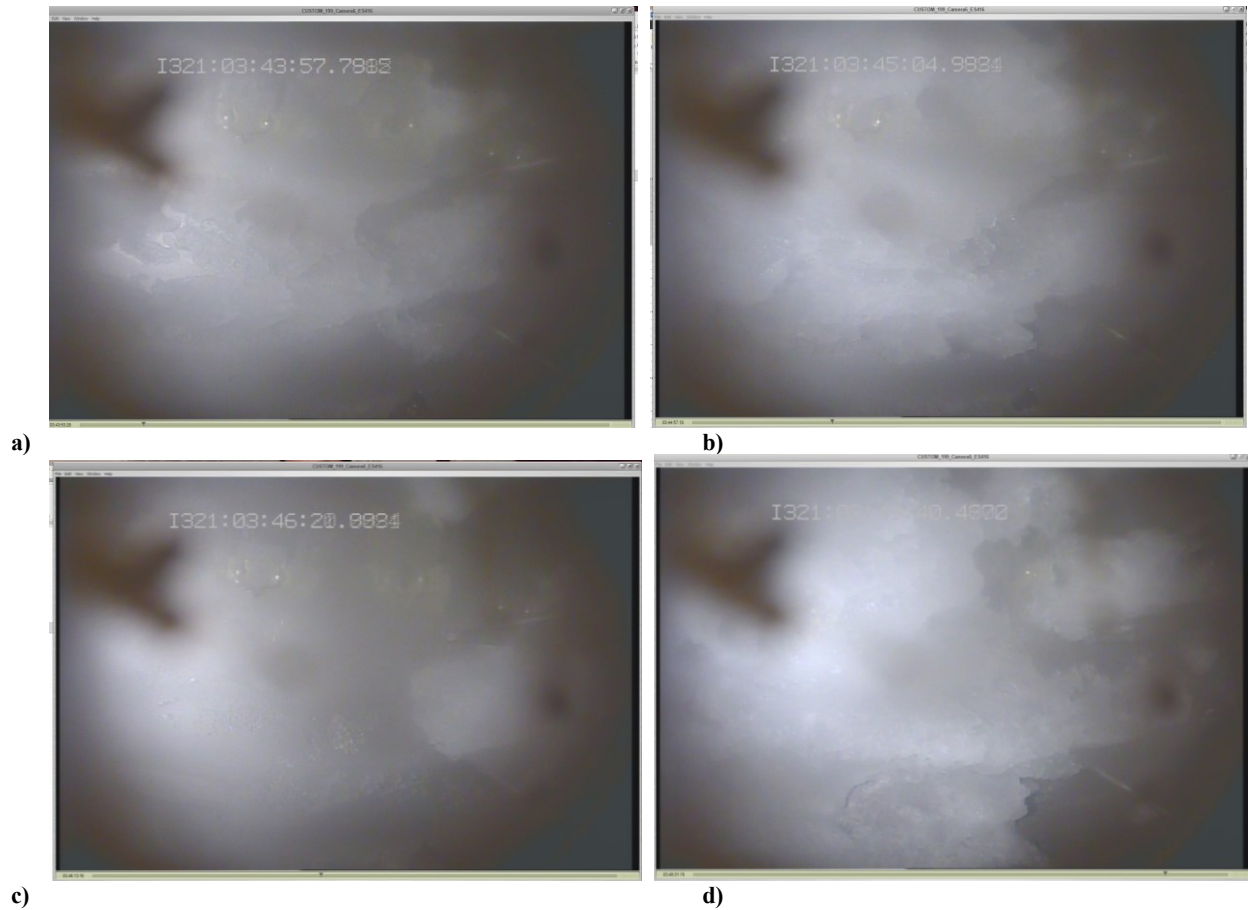
**Figure 20: Escort data point 093 a) Escort scan 72: 9.5% calculated blockage. b) Escort scan 150: 21% calculated blockage. c) Escort scan 208: 36% calculated blockage. d) Escort scan 305: 60% calculated blockage.**

The progression of blocked area versus Escort scan number that was calculated earlier and shown in Figure 16a) are corroborated by the images taken at discrete scans shown in Figure 20. The degree of area blockage that was calculate at Escort scan 72 is 9.5% and the corresponding image is shown in Figure 20a). The degree of area blockage that was calculate at Escort scan 150 is 21% and the corresponding image is shown in Figure 20b). Further, the degree of area blockage that was calculate at Escort scan 208 is 36% and the corresponding image is shown in Figure 20c). Similarly, the degree of area blockage that was calculate at Escort scan 305 is 60% and the corresponding image is shown in Figure 20d). Although there was no quantitative measurements taken of the magnitude of the ice buildup, it can be seen that the calculated values of area reduction are qualitatively confirmed. Likewise, based on the jagged surface features of the ice, it is reasonable to conclude that the loss increase versus Escort scan number that were estimated by the compressor flow model and are shown in Figure 18a), are qualitatively correct.

### D. Images at EGV stator 2 trailing edge during called engine roll back at low altitude (4800 ft.)

One of the NASA objectives was to reproduce an engine roll back at a significantly lower altitude, in order to see if the “icing criteria” that was observed during the high altitude roll back data points also applies at lower altitudes. This was done with the goal of providing an option in the future to test turbofan engines for ice crystal icing near sea level

conditions (without the need for an altitude chamber) at normal ambient temperatures observed in the winter. The PSL facility lower altitude limit is 4,800 feet. Escort data point 424 achieved a called engine roll back in 138 seconds, Several data points were taken at this lower altitude (wet bulb temperature plotted in Figure 7). The icing images for discreet scan numbers from Escort data point 416 are illustrated in Figure 21.



**Figure 21: Escort data point 416; a) Escort scan 86. b) Escort scan 153. c) Escort scan 229. d) Escort scan 429.**

The characteristic of the called roll back at low altitude was at reduced levels of wet bulb temperatures, as also seen in Figure 6. The fastest called roll back was 138 seconds after ice cloud was turned on. This is slower than the fastest called roll back points that were observed at the higher altitudes near 30,000 feet. Although the images appear to be cloudy, it is evident that there is ice buildup. The images in Figure 21 (Escort data point 416), illustrate one of the slower roll back points that was called at Escort scan 429. Ice buildup and shedding was observed as illustrated in Figure 21 b) and c). This ice buildup and shedding process continued through 393 seconds, where the engine roll back criteria was met.

### III. Conclusions

A turbofan engine known to have experienced an icing event at high altitudes and operating conditions during flight through convective ice crystal clouds was tested in the NASA Propulsion Systems Laboratory (PSL) simulated altitude engine test facility. The NPSS system model with parameter degradation of the LPC closely matched the measured engine performance of the full engine roll back data point.

The rate of blockage growth due to ice accretion is a function of the local wet bulb temperature as well as the ratio of ice-water-flow-rate to air-flow-rate (*IWAR*). The minimum value *IWAR* for engine roll back is on the order of 0.002.

For a given *IWAR*, the highest rate of (ice) blockage growth rate is a function of the wet bulb temperature at the EGV stator 2 trailing edge.

Based on this study, the upper limit of wet bulb temperature at stator 2 trailing edge is in the range of 493 - 497 R for ice accretion to occur. Likewise, the lower limit of wet bulb temperature at stator 2 trailing edge is in the range of 475 - 477 R in order for ice accretion to occur.

The video images of the EGV stator 2 trailing edge verify qualitatively that the area blockage which was estimated by compressor flow analysis can be attributed to ice accretion. However boundary layer growth is also a factor in the area blockage, but it has not been quantified.

Based on the data analysis results, a model was created, and consists of a general relation of the blockage growth versus the wet bulb temperature and the ratio of ice-water-flow-rate to-air-flow-rate (*IWAR*). The corresponding rate of pressure loss coefficient is related to the percent of blocked annular area.

During the full roll back point (Escort 093) there is a large increase in pressure loss coefficient in the tandem stator EGV, and is estimated to be mainly due to deteriorated surface roughness, and only minimally due to leading edge incidence loss effects. The temperatures at the EGV stator 2 region change significantly from their initial values prior to the ice cloud being turned on, yet the growth of blockage (partially due to ice accretion) continues unimpeded at the same rate through the engine roll back.

A low altitude engine called roll back was achieved at 4,800 feet in a similar time scale as those observed at high altitude (near 30,000 feet) at comparable values of *IWAR*. However, the wet bulb temperature to achieve accretion was lower than at the higher altitudes.

## Appendix

Escort	Roll Back,	Altitude	Flight Mach	Engine PT	Engine TT	TWC	Water /Air	Twbs	Twbs	Rel Hum	Melt Fraction	EGV2 10 s	Anti-Ice	Spinner
Data Pt.	sec			Inlet	Inlet, R	Engine Inlet	Flow Ratio	EGV1 LE	EGV2 TE	EGV2 TE	EGV2 TE	T metal, R		Heat
235	9999	27579	0.5598	6.0219	475.50	1.43	0.00312	491.60	500.58	9.6	0.1571	500.8	ON	ON
240	9999	27860	0.5547	5.9248	469.00	1.43	0.00311	487.20	496.80	9.1	0.0688	500.2	ON	OFF
200	382	28717	0.5681	5.7580	467.33	1.44	0.00325	486.60	496.31	8.7	0.0618	499.4	ON	ON
314	9999	27974	0.5586	5.9110	465.09	1.05	0.00229	485.20	495.10	9.1	0.0390	505.4	ON	ON
194	85	28482	0.5564	5.7688	462.41	1.41	0.00311	483.00	493.16	8.2	0.0119	498.6	ON	ON
464	9999	32093	0.5594	4.9030	466.73	0.79	0.00207	483.80	493.00	8.5	0.0136	506.8	ON	ON
178	77	28445	0.5544	5.7700	460.45	1.43	0.00315	482.00	492.30	8.5	0.0026	498.7	ON	ON
93	72	28947	0.5158	5.4891	459.50	1.36	0.00304	479.10	489.45	7.9	0.0000	497.0	ON	ON
462	209	32064	0.5587	4.9072	459.48	0.78	0.00201	479.50	489.37	7.6	0.0000	503.9	ON	ON
202	87	28486	0.5568	5.7694	454.29	1.43	0.00311	477.60	488.47	7.9	0.0000	496.9	ON	ON
344	186	31035	0.5578	5.1400	455.59	0.90	0.00219	477.40	487.84	7.1	0.0000	502.6	ON	ON
338	268	31033	0.5588	5.1465	447.24	0.92	0.00221	475.40	487.81	5.5	0.0000	503.4	ON	ON
444	219	32270	0.5713	4.9063	452.75	0.85	0.00218	475.80	486.31	7.1	0.0000	500.5	ON	ON
454	349	32068	0.5582	4.9043	452.30	0.77	0.00196	474.80	485.45	6.9	0.0000	501.4	ON	ON
332	285	31046	0.5588	5.1436	449.98	0.85	0.00205	473.60	484.67	6.6	0.0000	501.1	ON	ON
340	9999	31110	0.5623	5.1200	437.45	0.90	0.00210	464.90	477.29	5.4	0.0000	497.4	ON	ON
506	9999	30010	0.5585	5.3918	431.90	1.42	0.00314	462.10	475.20	5.8	0.0000	491.9	OFF	OFF
406	9999	4885.3	0.1925	12.6019	457.43	3.24	0.00270	476.80	489.04	10.6	0.0000	498.3	ON	ON
422	148	4882.7	0.1919	12.6013	463.92	4.31	0.00369	476.30	483.36	27.0	0.0000	494.2	ON	ON
424	138	4871.5	0.1901	12.6005	466.96	4.37	0.00378	477.40	483.31	33.2	0.0000	494.0	ON	ON
416	393	4862.5	0.1897	12.6033	449.00	3.81	0.00311	469.10	482.17	10.0	0.0000	494.9	ON	ON
418	210	4853.5	0.1886	12.6039	452.81	4.49	0.00371	471.30	482.12	14.3	0.0000	494.3	ON	ON

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