

# Development of the Space Debris Sensor (SDS)

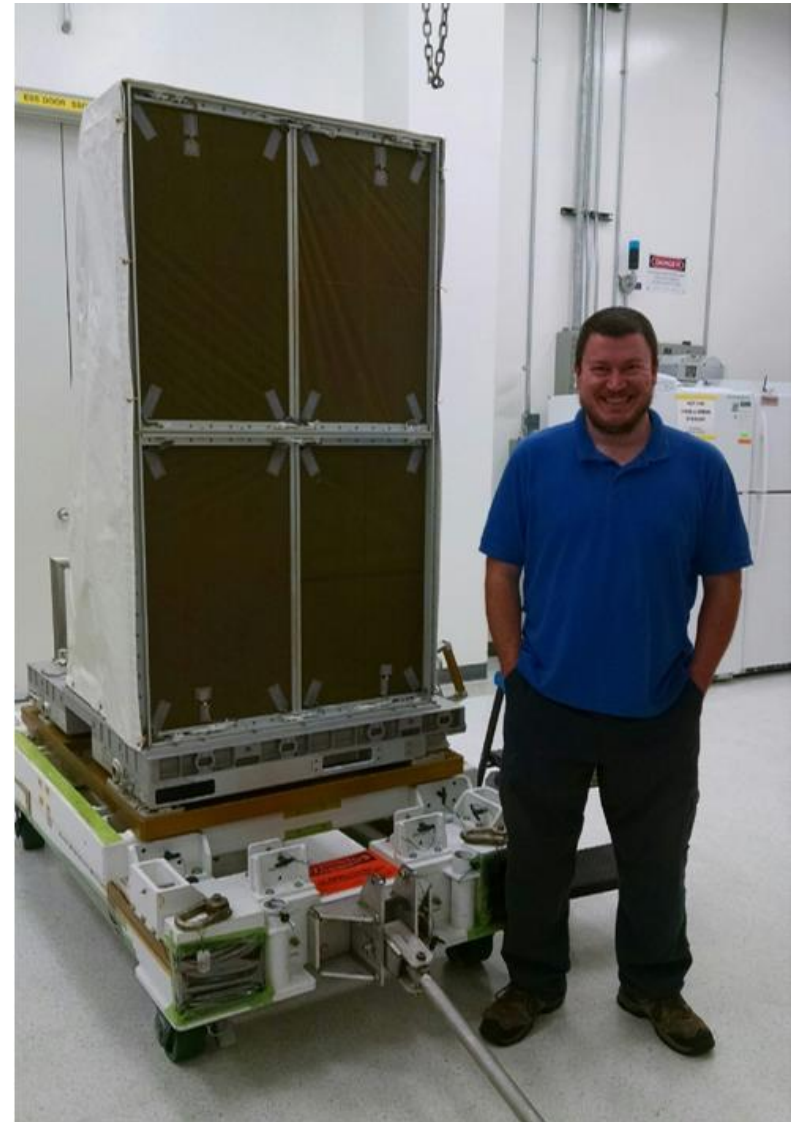


Joe Hamilton  
SDS Principal Investigator  
January 31. 2017



# Outline

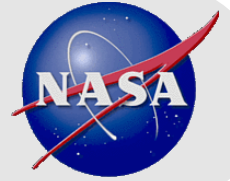
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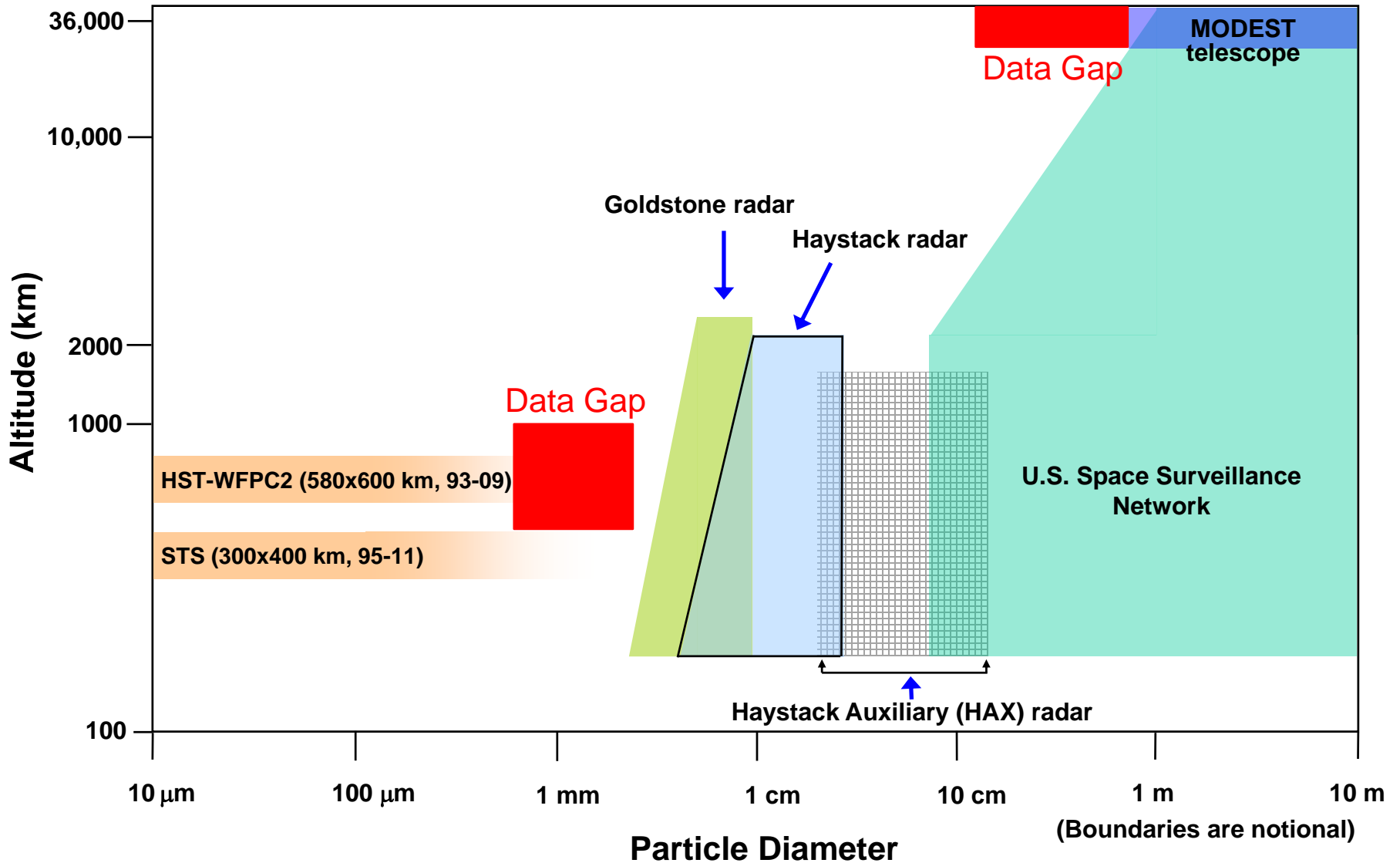


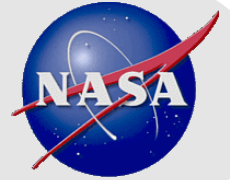
## Background

- DRAGONS concept and technology has been under development with intermittent grants since 2002
- The goal of DRAGONS is to provide in-situ statistical data on the debris population that is too small for ground-based remote sensing to accomplish.
  - Results would be used to update the Orbital Debris Engineering Model (ORDEM)
  - **Current estimate of the small debris population is based on inspection of exposed surfaces returned on Shuttle (Retired 2011)**
- The DRAGONS team includes the NASA Orbital Debris Program Office, the NASA Hypervelocity Impact Technology group, the NASA/JSC Engineering Directorate, Jacobs, the United States Naval Academy, the Naval Research Lab, Virginia Tech, and the University of Kent.



# Orbital Debris Measurement Coverage





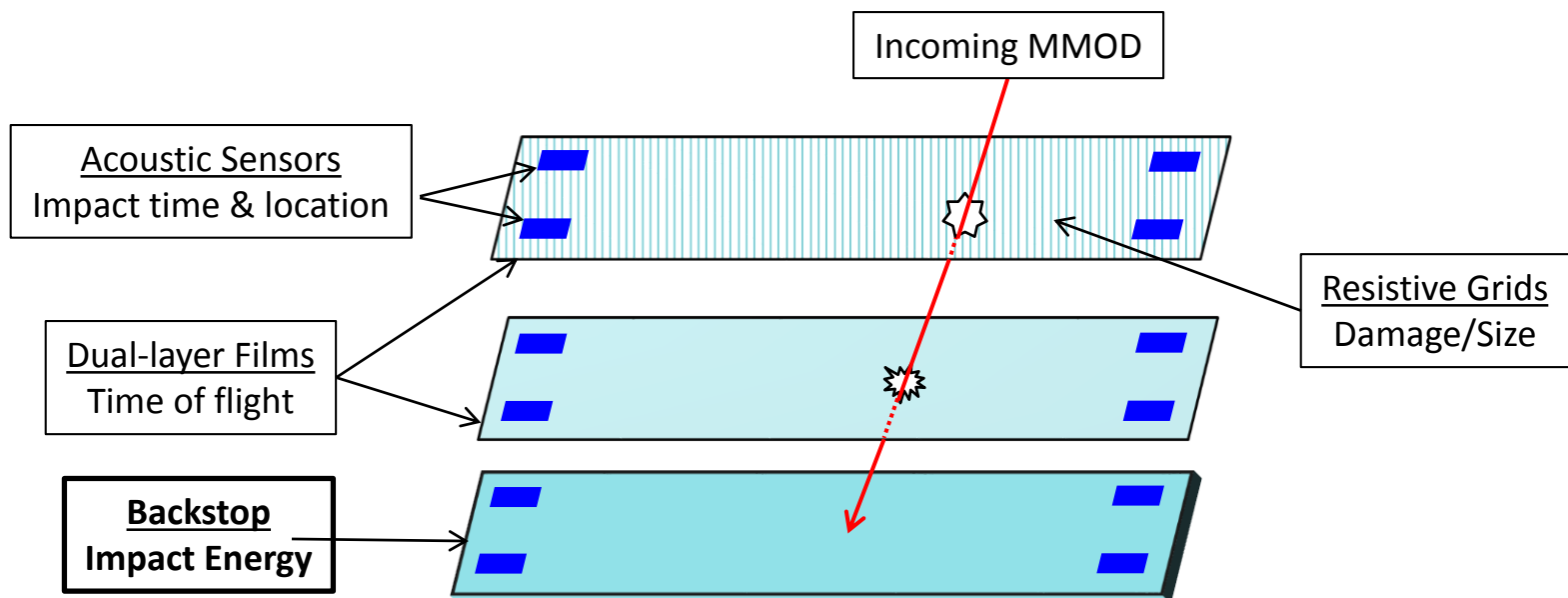
## What is SDS?

- **DRAGONS is an impact sensor designed to detect and characterize collisions with small orbital debris.**
  - 50 $\mu$ m to > 1mm debris size detection
  - Characterize debris size, speed, direction, and density
- **The Space Debris Sensor (SDS) is a flight demonstration of DRAGONS on the International Space Station**
  - Approximately 1 m<sup>2</sup> of detection area facing the ISS velocity vector
  - Minimum two year mission on Columbus External Payloads Facility (EPF)
  - Minimal obstruction from ISS hardware
  - Development is nearing final checkout and integration with the ISS
  - Current launch schedule is SpaceX 13, ~ Sept 2017, or SpaceX 14, ~ Jan 2018

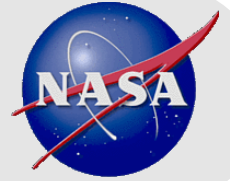


# Detection Principles

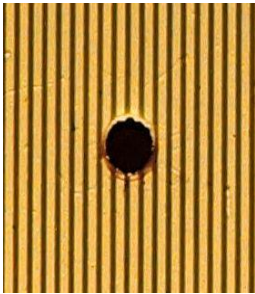
- SDS combines dual-layer thin films, an acoustic sensor system, a resistive grid sensor system, and sensed backstop
- Impact detection and recording capability
  - **Impact time, particle size, impact speed, impact direction, and impact energy/particle density**



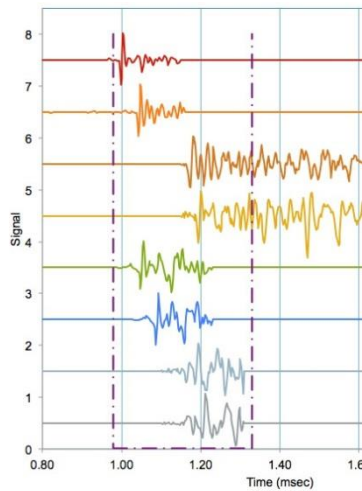
# Example 0.4 mm 30° Stainless Steel 7 km/s



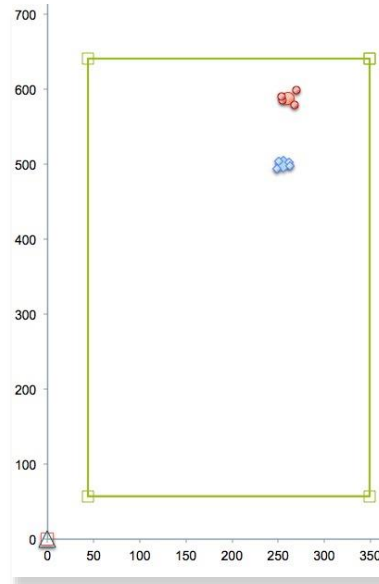
Layer 1 hole  
Broke 3 lines



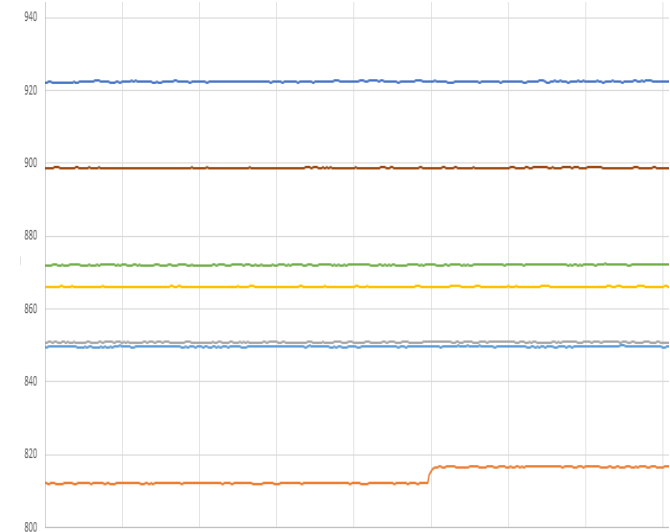
Filtered Acoustic Data



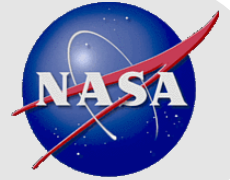
Location  
Red=Layer 1;  
Blue = Layer 2



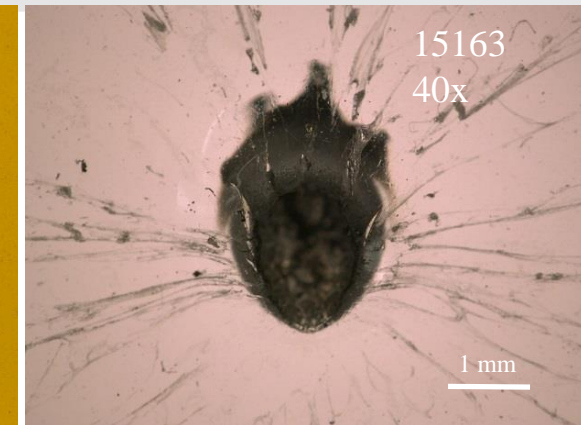
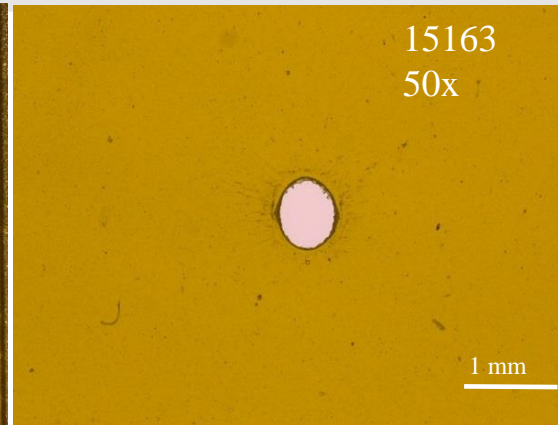
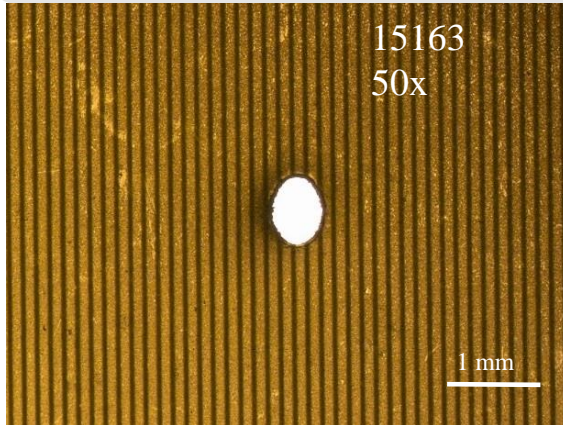
Resistance change  
consistent with 3  
line break



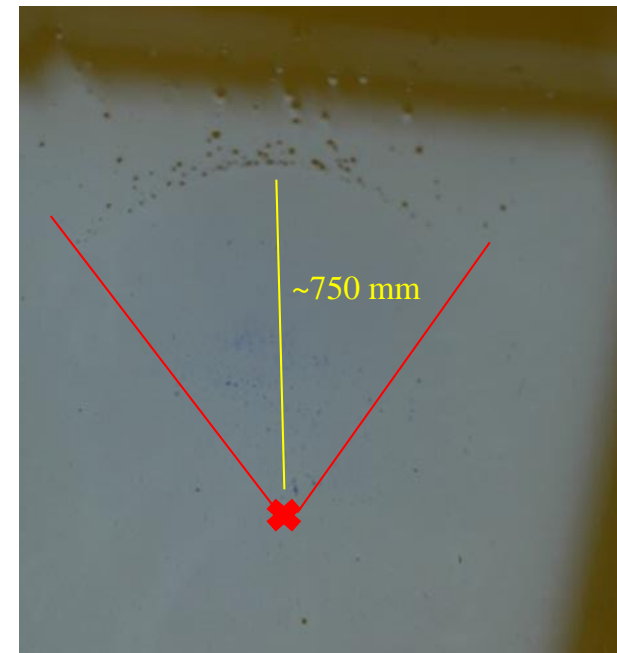




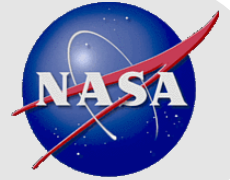
# 500 $\mu$ m 440C Stainless Steel



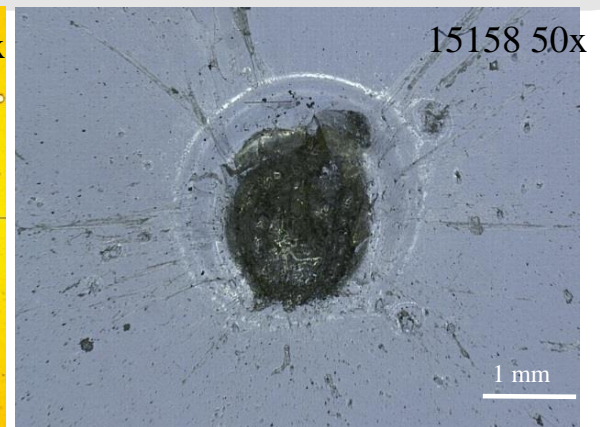
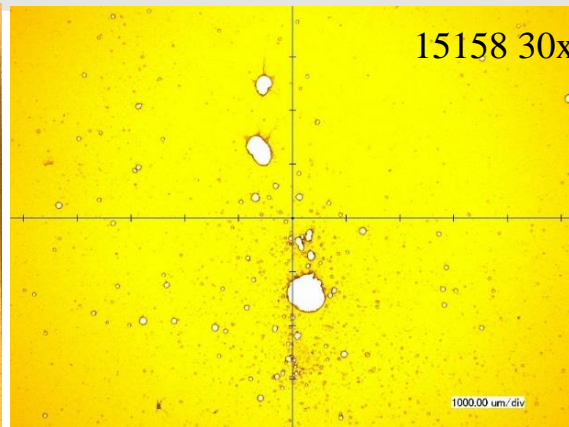
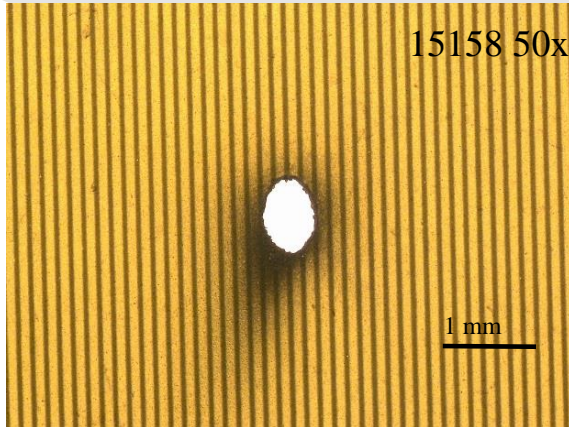
- Steel maintains shape throughout
- No visible break up of particles during impacts
- Steel shots produce significant secondary ejecta from Lexan back plate
- Ejecta has enough velocity to penetrate and dimple Kapton layer in wide arc downstream from shot
- Straight-on shots produce halo around entry hole; As shot angle increases the damage moves further away from hole



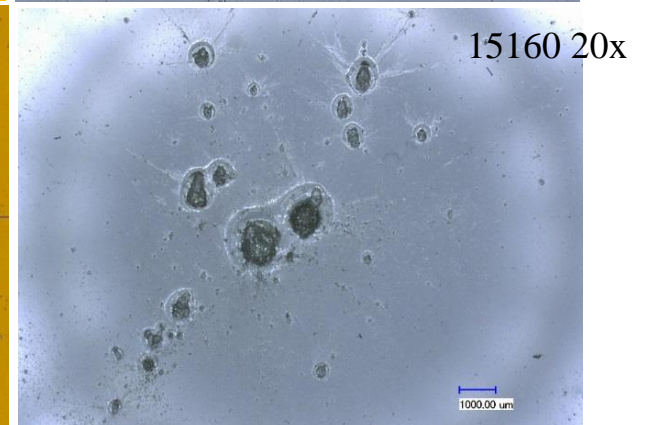
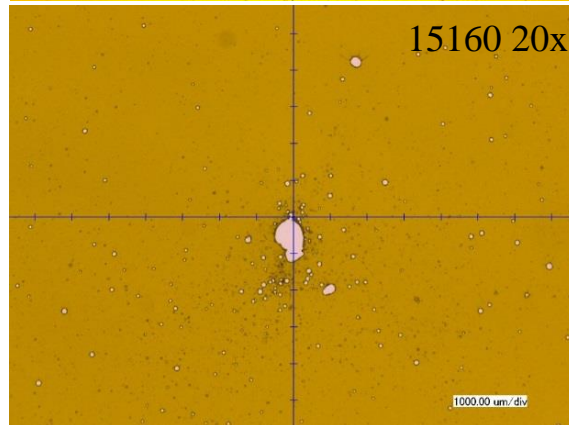




# 500 $\mu$ m Aluminum Al 2017-T4

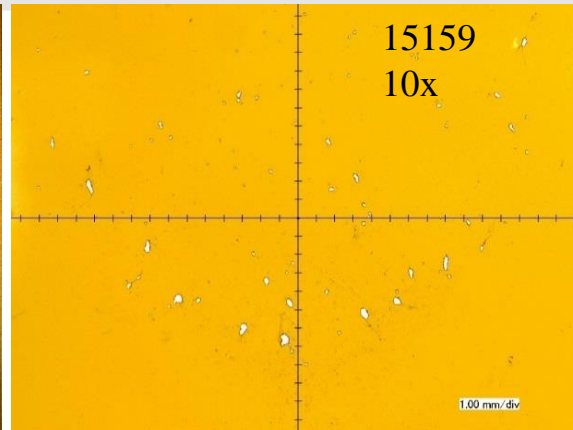
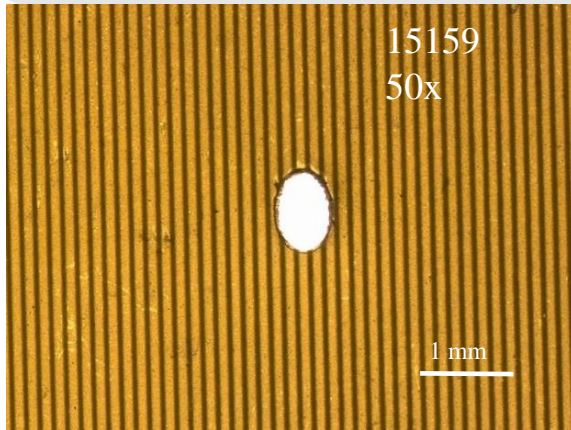


- Aluminum particles show break-up after first layer
- Amount of break up varied in the three shots
- One shot left a clean crater on Lexan back plate
- Two other shots had a collection of smaller craters
- No sign of ejecta damage on Kapton layer

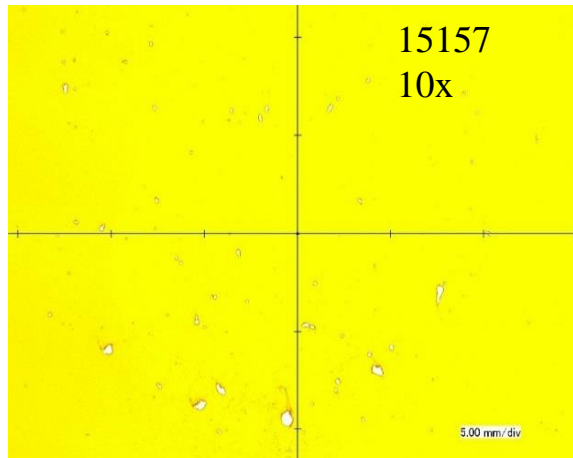




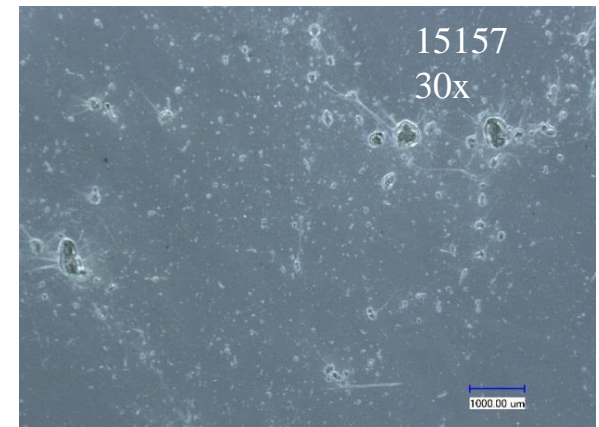
# 500 $\mu$ m PMMA Plexiglass

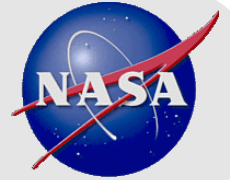


- Plastic particles broke up significantly after impacting the NCAS grid
- ‘Half circle’ hole pattern on Kapton layer with largest hole at bottom of the circle
- Same break up pattern for all shots - ~25mm wide by 20mm tall area of holes

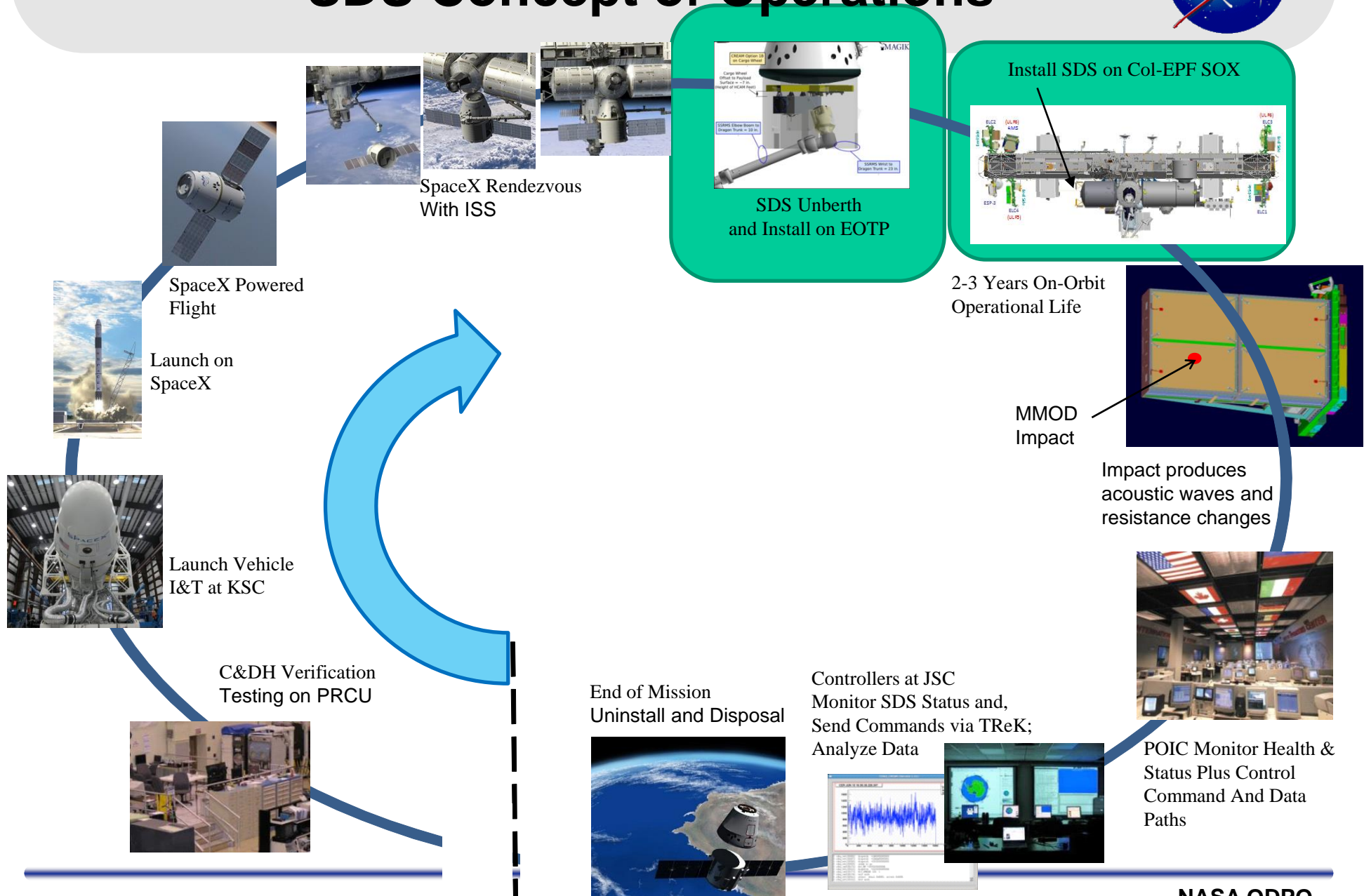


- Only one plastic shot showed up on the Lexan back plate
- No craters on Lexan – only residue





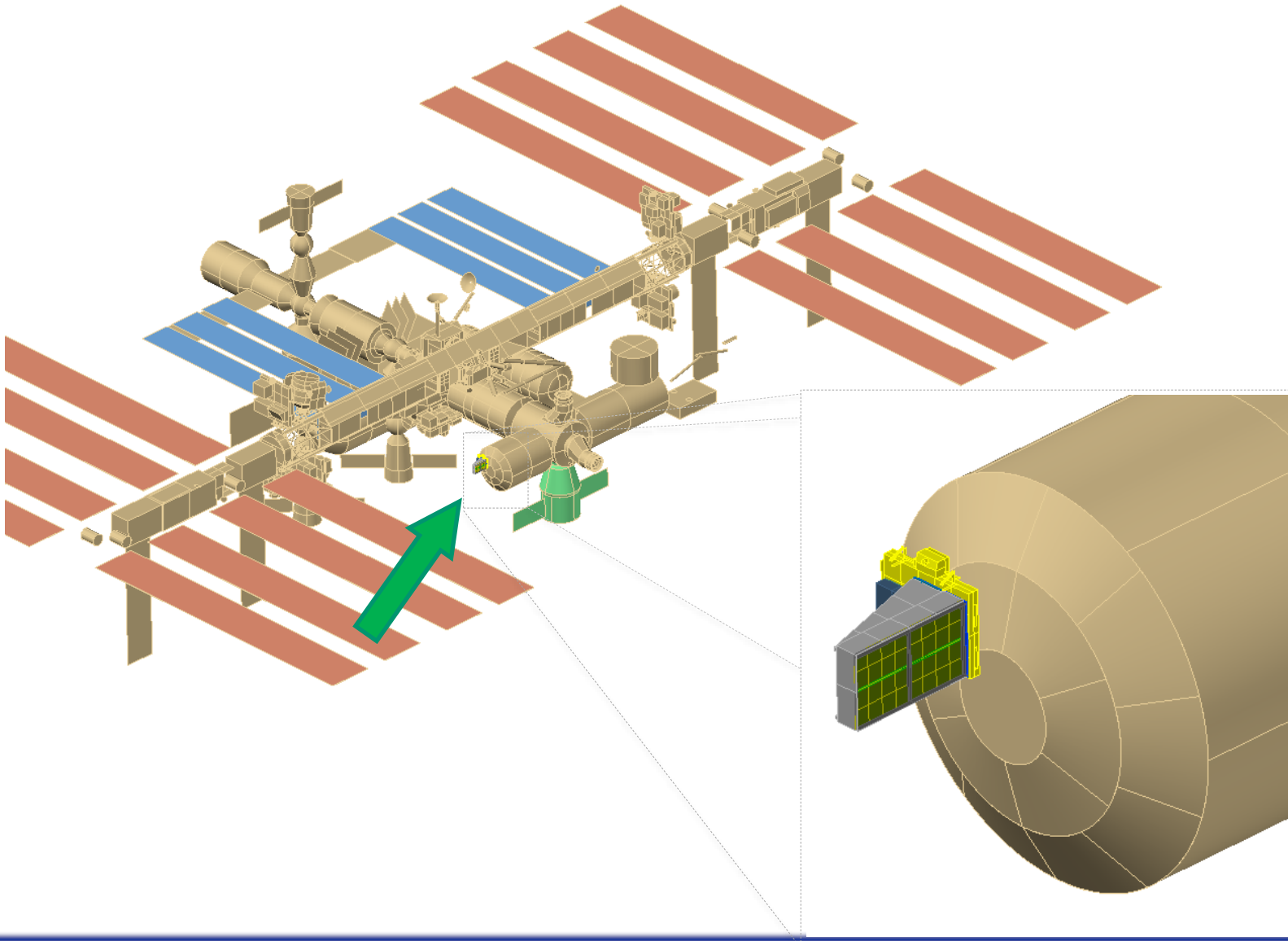
# SDS Concept of Operations





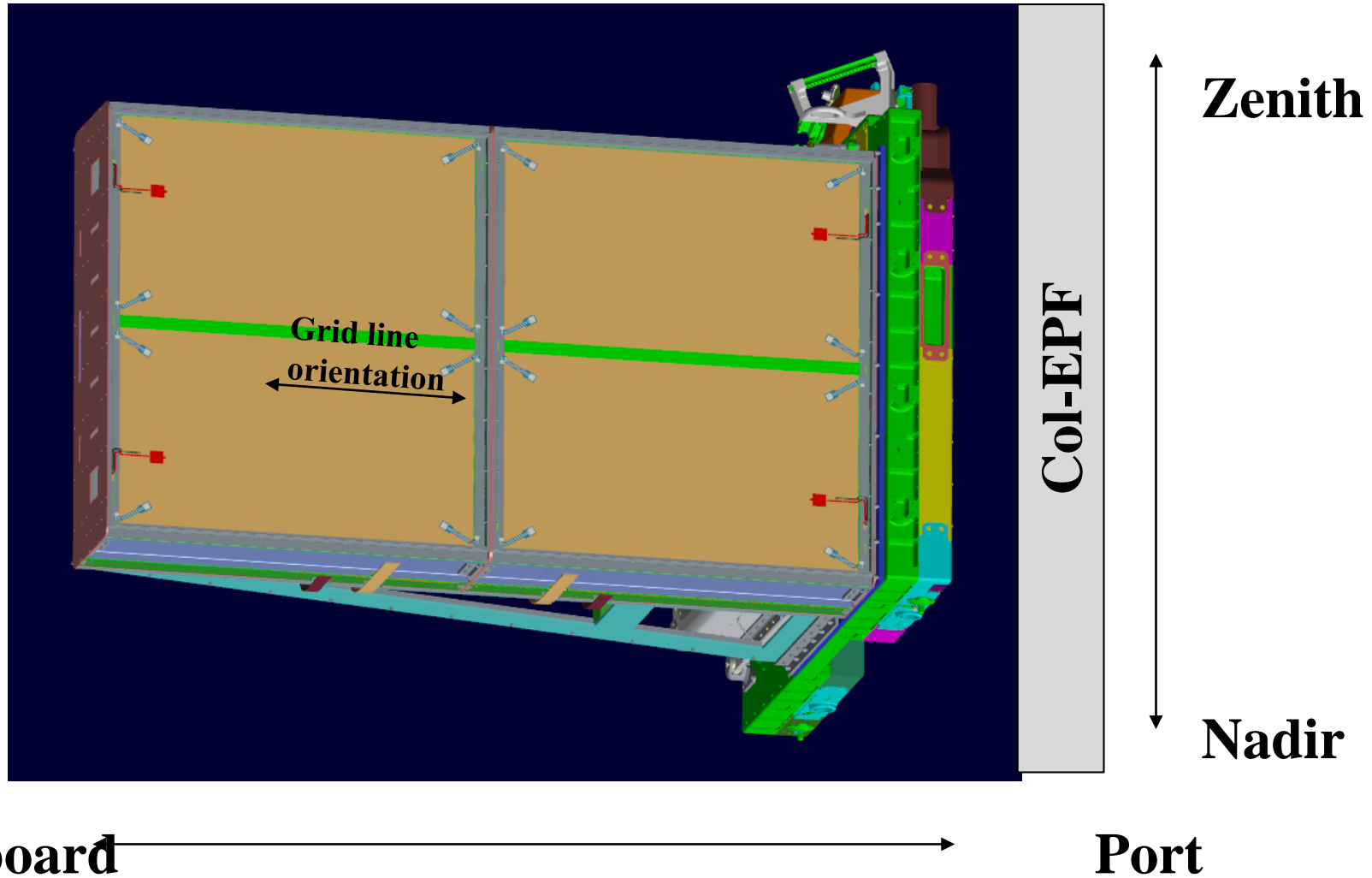


# SDS on Columbus-External Payload Facility





# SDS ISS Orientation



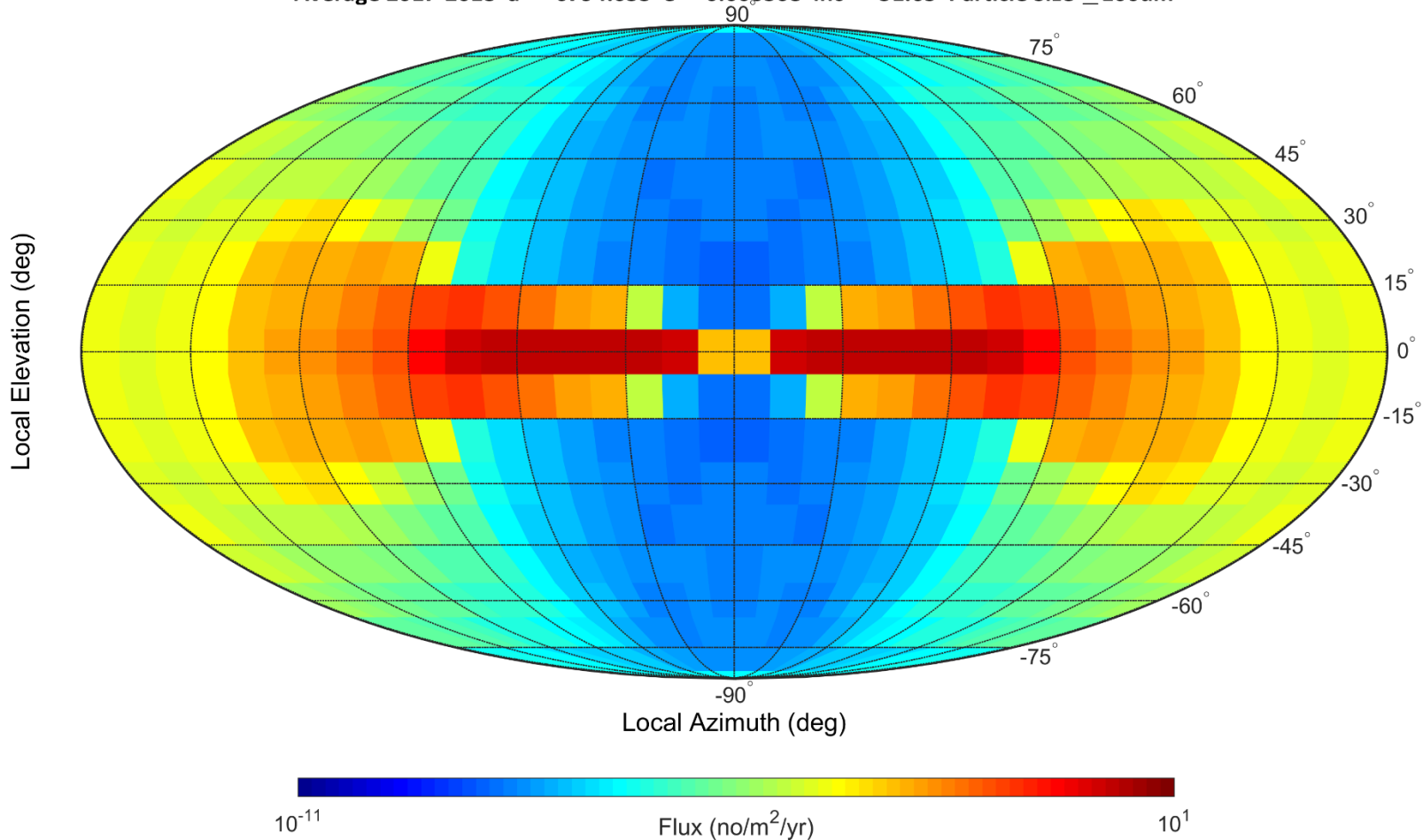


# 2-D Directional Flux – ORDEM 3.0

## Orbital Debris Engineering Model (ORDEM 3.0)

2-D Directional Flux

Average 2017-2018  $a = 6794.635$   $e = 0.000368$   $inc = 51.65$  Particle Size  $\geq 100\mu m$





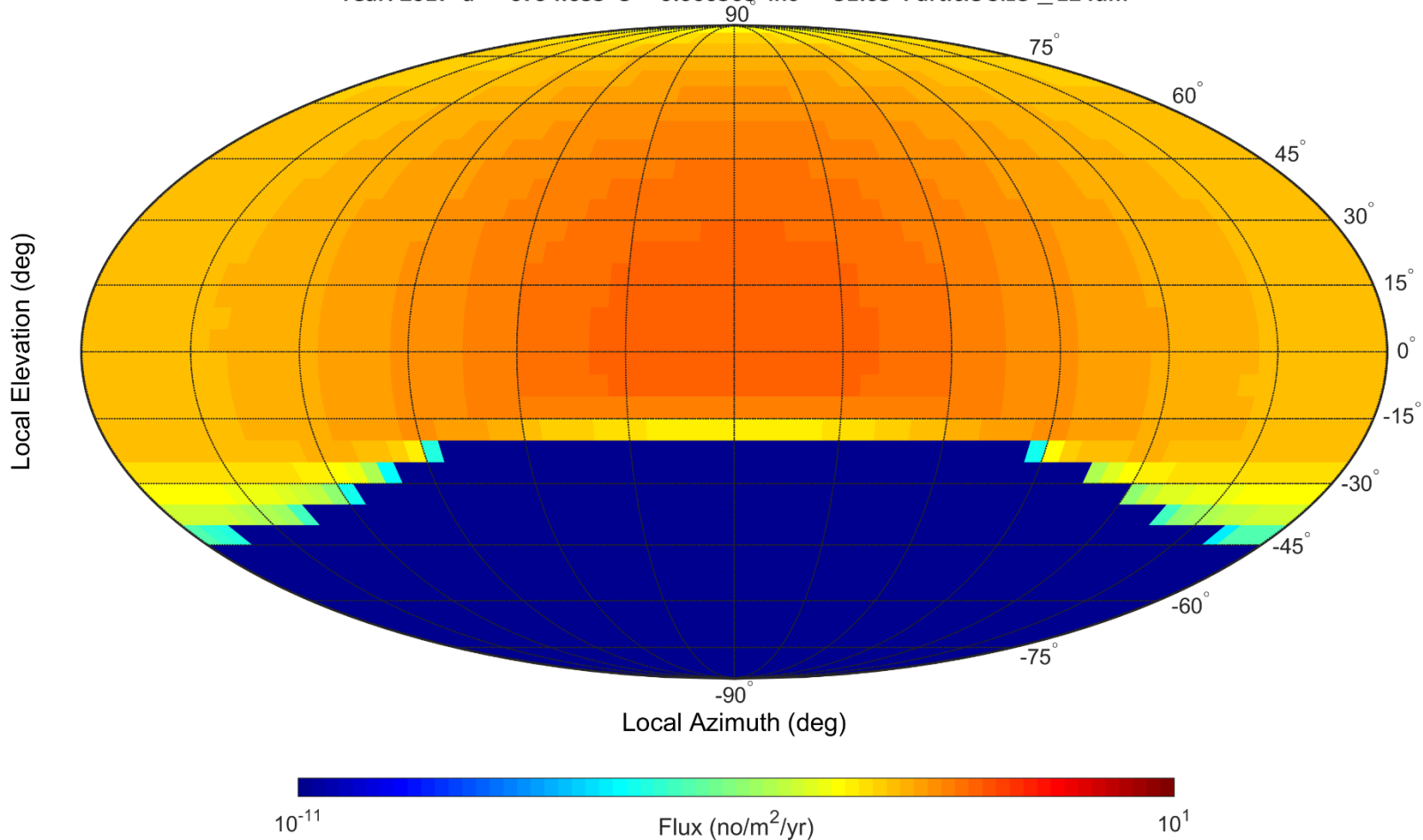


# 2-D Directional Flux – MEM 2.0

## Micro-Meteoroid Engineering Model (MEM 2.0)

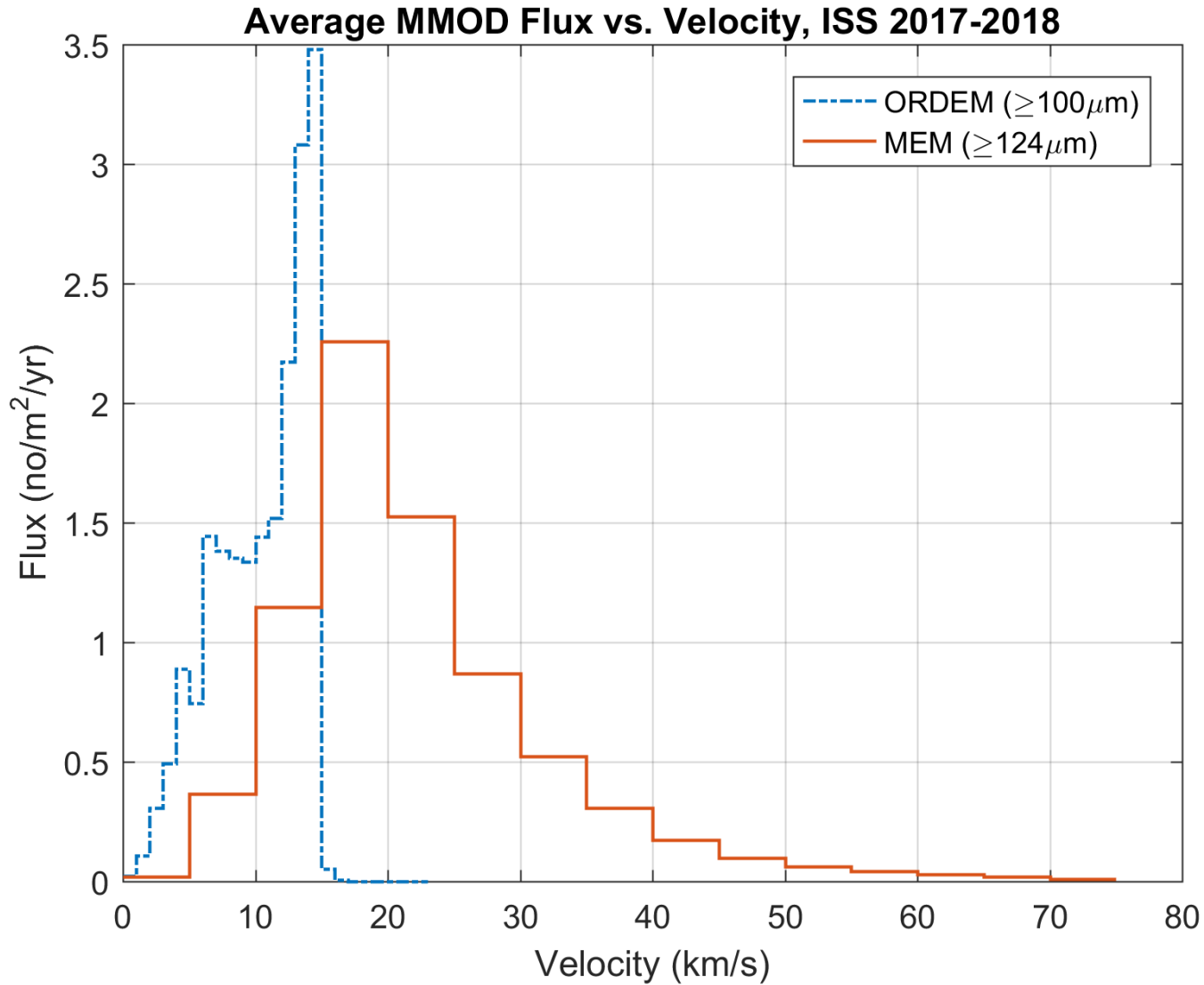
2-D Directional Flux

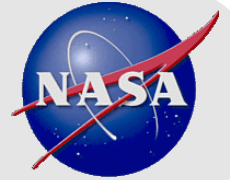
Year: 2017 a = 6794.635 e = 0.000368 inc = 51.65 Particle Size  $\geq 124\mu\text{m}$





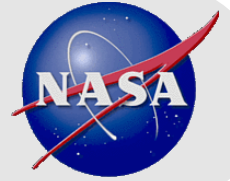
# Predicted Flux vs. Velocity





## Conclusions

- **SDS is the top priority for NASA ODPO development of orbital debris monitoring capability**
  - Addressing a gap in detection coverage
  - SDS will inform the design of future DRAGONS
- **The NASA ODPO will use the experience from SDS to improve the detection and characterization technology.**
  - Improved grids with 50 $\mu$ m width lines
  - Larger detection areas
  - Improved acoustic algorithms for speed, direction, and density calculations
- **The NASA ODPO is pursuing additional flight opportunities to put DRAGONS at higher altitudes**
  - Targeting flights in the 700 to 1000 km altitude region
  - Sun-synchronous orbits



# Questions?

