



Revisiting the Birth of 7YSZ Thermal Barrier Coatings: Steve Stecura

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Abstract

Thermal barrier coatings are widely used in all turbine engines, typically using a 7 wt% Y_2O_3 - ZrO_2 formulation. Extensive research and development over many decades have refined the processing and structure of these coatings for increased durability and reliability. New compositions demonstrate some unique advantages and are gaining in application. However, the “7YSZ” formulation predominates and is still in widespread use. This special composition has been universally found to produce nanoscale precipitates of metastable t' tetragonal phase, giving rise to a unique toughening mechanism via ferro-elastic switching under stress. This note recalls the original study that identified superior properties of 6 to 8 wt% YSZ plasma sprayed thermal barrier coatings, published in 1978. The impact of this discovery, arguably, continues in some form to this day. At one point, 7YSZ thermal barrier coatings were used in every new aircraft and ground power turbine engine produced worldwide. It is a tribute to its inventor, Dr. Stephan J. Stecura, NASA retiree.

Background

Yttria stabilized zirconia (YSZ) thermal barrier coatings (TBC) are widely used in turbine engines to protect underlying metal structures from the intense heat of combusted jet fuel. The standard industry composition is 7 wt% Y_2O_3 - ZrO_2 . It has been estimated that industry capital expenditures for finished YSZ thermal barrier aerospace coatings reached \$1500 M, just for 2015 alone,¹ with 1 to 2 Mkg of YSZ used for plasma sprayed coatings (Ref. 1). Indeed, a Google search on the exact phrase “YSZ Thermal Barrier Coatings for Aircraft Engines” provided 35,800 hits from all sources, indicating the high level of general interest.

The early years, around 1970 and before, saw increasing interest and trials of air plasma sprayed (APS) oxides and fully stabilized zirconia (~12 wt% Y_2O_3) as thermal barriers for hot components in general (Refs. 2 to 4). However use in critical aircraft components was restricted because of limited cyclic lifetime: the coatings had a propensity to spall from compressive thermal stress, aggravated by the oxidation of underlying substrates or bond coats.

Seminal Experimental Findings

Fully stabilized zirconia, using MgO, CaO, or Y_2O_3 to stabilize the cubic phase of ZrO_2 , had become fairly standard practice then. Y_2O_3 provided more long term durability than the other stabilizers which tended to volatilize, hydrate, and destabilize the cubic phase. Compositional studies of YSZ were minimal and simply recommended high Y_2O_3 cubic phases. However one enabling discovery for present day TBCs came when partially stabilized (tetragonal) zirconia was first identified as more resistant to thermal

¹S. Nichani, K. Pillai, J. Varma. Unpublished market research, Grand View Research, www.grandviewresearch.com

cycling than fully stabilized cubic zirconia (1978) (Refs. 5 and 6). The kernel of the results from an extensive plasma spray TBC system optimization (Dr. Stephan Stecura, NASA TM 78976, 1978) is provided in Figure 1. The optimal 6 to 8 wt% YSZ composition was confirmed in furnace, natural gas/oxygen torch, and Mach 1 jet fueled burner rig tests (using actual air cooled PWA J75 (JT4A) turbine blades), totaling about 75 sample conditions overall, most with duplicates. This screening study was one in a series of developmental TBC studies (Ref. 7).

Pertinent conclusions were summarized as: “The data indicate that the best thermal barrier systems consisted of combinations involving the Ni-16.4Cr-5.1Al-0.15Y and Ni-17.0Cr-5.4Al-0.35Y bond coatings and the 6.2 Y₂O₃-ZrO₂ and 7.9 Y₂O₃-ZrO₂ stabilized zirconium oxide layers (all in wt%).” Furthermore, testifying to the durability on air-cooled blades: “...at 1580 and 1550 °C surface temperatures, withstood 1300 and 1500 1-h cycles without failure, respectively. In the Mach 1.0 burner rig the above two systems withstood over 1400 and 2000 1-h-cycles without failure at about 1480 and 1470 °C surface temperatures, respectively.”

Similar trends were confirmed in a subsequent optimization study, as illustrated in Figure 2. Here coated coupons were tested in cyclic furnace tests until the first visible crack in the TBC. The bond coat for this series was ~Ni-16Cr-6Al-0.17Y and achieved a maximum life near 400 cycles. Greater lives approaching 1400 cycles were produced with another bond coat having a ~Ni-35Cr-6Al-1.0Y composition, with presumably a lower CTE because of α -Cr precipitates. The success of YSZ coatings was highly leveraged by numerous concurrent plasma sprayed TBC studies at NASA Glenn (formerly Lewis) Research Center, most notably component and engine tests championed by Curt Liebert as far back as 1972 (Refs. 1, 2, and 6 to 9). No comparable studies of bulk Y₂O₃-ZrO₂ properties existed at the time, although the phase diagram of Scott did identify the metastable tetragonal phase in that range.

Growing Interest and Impact

An indication of the level of world-wide interest in yttria-stabilized zirconia YSZ thermal barrier coatings can be surmised from internet searches of journal articles using Google Scholar. In that 1970 to 1985 time frame, there were only about 18 related articles in standard materials science journals that followed YSZ TBC research, Table 1. They were predominately in Thin Solid Films. In the next 5 years Surface and Coatings Technology led with 19 articles. Finally, over the last 25 years, YSZ coating articles in this one journal alone burgeoned to nearly 650. Contributions in the journals listed totaled ~2500 YSZ coating articles in that time frame. A simple Google search yielded about 7000 hits from all sources (bottom row). The early “yttria-stabilized zirconia thermal barrier coating” items outside the journals were primarily NASA (<https://www.sti.nasa.gov/>) and other government agency reports or society proceedings. Clearly the YSZ thermal barrier industry has grown exponentially, with most contributions after 1990 and the successful ~7YSZ composition as the most prevalent composition. A case can be made that the success of optimal 7YSZ TBC coatings led to that composition being adopted for other applications, such as oxygen conductors in sensors or fuel cells.

More Compositional Details and Overall Context

An overview of the mechanisms, complexity, and critical refinements to achieve this growth can be appreciated from a number of key contributions and reviews, offered chronologically in the closing references (Refs. 4, 11 to 18). Many of these and current topics of high interest have been covered in the 2012 topical issue of the MRS Bulletin (Refs. 1, 19 to 22). While these represent only a fraction of the pertinent literature, virtually all refer to “7YSZ” or “6 to 8 wt% Y₂O₃-ZrO₂” as the standard TBC topcoat composition, essentially unchanged from its original discovery in 1978.

The attributes of the 1978 study believed to be key to identifying the preferred 6 to 8 wt% YSZ performance were experimental thoroughness and process control for what was then basically a hand spray process. The consistent life data is a tribute to the long-time skilled APS operator, Jack E. Brown.

Consistent bond coat and top coat thicknesses were obtained, necessitating frequent measurements during processing. Care was taken to apply APS bond and top coats within minutes of pre-conditioning by a light grit blasting to improve bonding and avoid uncontrolled random variations by ambient humidity and native oxide effects.

The composition of the Ni-16Cr-5Al-Y bond coat represented the then current range for optimal alumina formation and ductility consistent with results on bulk alloys provided by the oxidation literature at that time. The Y-dopant level was varied at 0.15, 0.35, 0.61, and 1.08 wt%. Initially, the best lives were obtained for Ni-16Cr-5Al at the 0.15 and 0.35 wt% Y levels, which allowed the full cyclic life of the TBC to be realized. High Y levels are generally needed in air plasma spray compared to PVD or bulk alloys because some of the Y is oxidized during spraying, thus diminishing Y in solution. Too high a level results in YNi₉ precipitates and disruptive oxidation behavior in those regions. None of the more robust alumina scale adhesion theories accepted today were even proposed at that time, but empirical results had indicated this strong sensitivity to low levels of Y dopants.

The Y₂O₃ content of the ZrO₂ was varied as 4.0, 6.2, 7.9, 11.5, 17.4, and 24.4 wt% Y₂O₃-ZrO₂. At that time, the t' metastable tetragonal was not identified in YSZ coatings until Miller (1981), so there was no specific rationale for the optimal behavior (except avoidance of the metastable tetragonal t-phase in lower Y coatings that does transform to monoclinic after cycling). It is now well known that the optimal behavior of 7YSZ derives from the formation of the t' phase that does not transform martensitically to the disruptive monoclinic ZrO₂ phase on cooling. Nor is it susceptible to the transformation toughening that had been popular at the time in coining the term “ceramic steel.” It does not transform upon deformation and is so termed “non-transformable” tetragonal, at least with respect to the monoclinic daughter phase, as originally termed in Miller, et al. (Ref. 6). Initially, the durability of 7YSZ corresponds to this slightly tetragonal t' phase, as fine precipitates having three possible orientation relationships to the parent cubic (Refs. 6, 23 to 25). Ultimately it was shown that ferro-elastic switching on a nanoscale can explain the toughening mechanism for 7YSZ and its superior performance (Refs. 26 and 27). The failure locus was originally identified as a crack in the YSZ layer near the bond coat, essentially the same as current APS failure descriptions.

The Aftermath: Processing, Understanding, and the Future

These features, i.e., the t' YSZ phase and an oxidation resistant MCrAlY bond coat, provided the starting point for highly durable coatings. Stecura's additional improvements to these early APS coatings revolved around high 25 to 35 wt% Cr, Yb-doped MCrAl bond coats (Ref. 28) and 12 to 15 wt% Yb₂O₃-ZrO₂ top coats (Ref. 29). However the interest and improvements in 7YSZ based TBC systems continued to this day. Most notable contributions would be EB-PVD (electron beam-physical vapor deposition) coating processing for strain tolerance and aerodynamically smooth airfoil surfaces. Also, robotic APS topcoat deposition, LPPS MCrAlY, and CVD processing of Ni(Pt)Al bond coats have all improved durability and increased use. Oxidative failure mechanisms have enhanced understanding, with finite element models showing CTE mismatch stress concentrations at asperities in plasma sprayed MCrAlY. Oxidatively ratcheted alumina scales (thermal grown oxide, TGO) and ridged or rumped Ni(Pt)Al bond coats cause stress concentrations and failure upon cycling. Thermal gradients and sintering of the YSZ have been used to refine failure models, especially in high heat flux laser tests.

Compositionally, low SiO₂, high purity YSZ is an initial prerequisite to minimize sintering and loss of compliance. Small particle plasma spray (SPS) and solution precursor plasma spray (SPPS) offer unique advantages in tailoring strain tolerant, low k TBC microstructure or uniform composition of chemical modifications. Multiply doped, low k, sinter- and CMAS-resistant YbGd-doped cubic YSZ and (La or Gd)₂Zr₂O₇ pyrochlores represent successful new directions in commercial TBC chemistry beyond 7YSZ (Ref. 30). All the advances mentioned above are covered in the aforementioned reviews and the details are beyond the scope of this simple memorandum.

Conclusion

Nevertheless the current dominance of the 7YSZ chemistry for coatings already in service can be directly traced to the original studies by Stecura. It is therefore a remarkable historical point that this empirically derived coating composition endures some 35 years later, with arguably a direct impact on current technology. Compared to other continually evolving turbine related material compositions over the same time frame, such as directionally solidified or single crystal nickel-base superalloys (gen 1–5), doped Pt-aluminide (+Hf, Si, etc.) bond coats and dual microstructure disk alloys, 7YSZ appears relatively unchanged from this 1978 study. 7YSZ has become so widely used now that the impact of the original finding, often lost in a background search, cannot be overemphasized and is a credit to its inventor. It is important to recognize that the discovery, prior to numerous subsequent advances in characterization and understanding, was accomplished by structured, well-informed empiricism coupled with experimental diligence.

Dr. Stephan Stecura emigrated to the United States from Ukraine, worked on high temperature materials at NASA Glenn Research Center and retired in 1994, and lives in the Cleveland area. He was well known for his discerning, enigmatic approach to research, with a smile on his face and treasure troves of original data tucked away in notebooks. The authors are fortunate to have witnessed and studied his successes.

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TABLE 1.—GOOGLE SCHOLAR HITS FOR ALL WORDS (YSZ THERMAL BARRIER COATINGS) ANYWHERE IN JOURNAL ARTICLE. LAST ROW INDICATES HITS FOR ANY SOURCE USING SEARCH TERM YSZ AND EXACT TERM “THERMAL BARRIER COATING”

Google Scholar Searches	Term: YSZ thermal barrier coatings		
	1970 to 1985	1985 to 1990	1990 to 2016
Surface and Coatings Technology	0	19	648
J. Thermal Spray Technology	0	0	483
J. American Ceramic Society	0	0	429
Ceramics International	1	0	355
J. European Ceramic Society	0	0	285
Thin Solid Films	17	8	99
Acta mat	0	0	146
Scripta mat	0	0	48
(select journals) total	18	27	2493
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all sources (+ patents)	181	183	6940

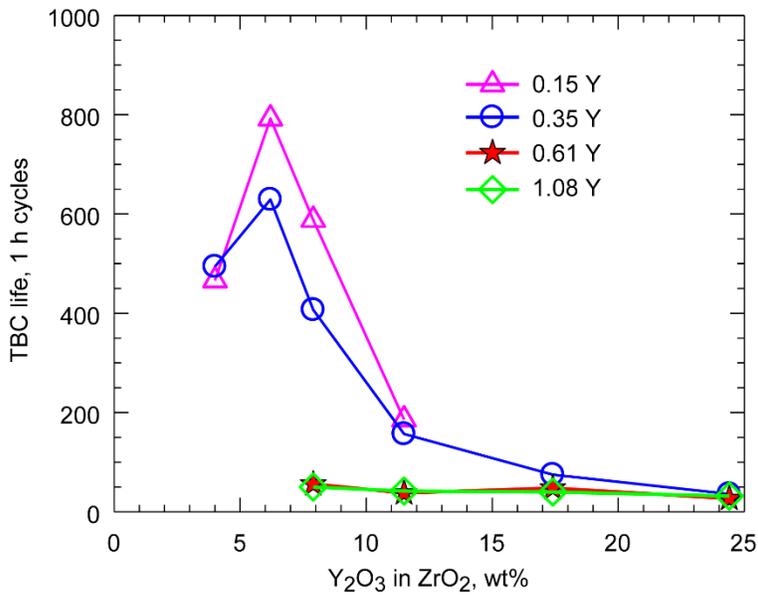


Figure 1.—TBC life (time to first TBC crack) in 1200 °C 1-h cyclic natural gas-O₂ torch rig test of APS YSZ as a function of Y₂O₃ content in YSZ, using Ni-16Cr-5Al-Y bond coats. From S. Stecura, “Effects of Compositional Changes on the Performance of a Thermal Barrier Coating System,” NASA TM 78976, (1978). (also presented in Darolia, 2013).

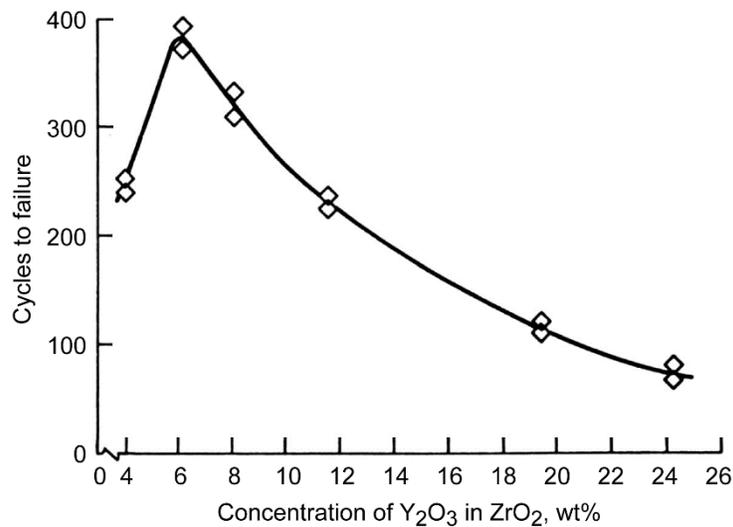


Figure 2.—TBC life (time to first TBC crack) in 1110 °C 1-h cyclic furnace test as a function of Y₂O₃ content in YSZ; Ni-16Cr-6Al-0.2Y bondcoat. From S. Stecura, “Optimization of the NiCrAlY/ZrO₂-Y₂O₃ Thermal Barrier,” NASA TM-86905, 1985 (also presented in Miller, 2009).

