International Space Station Lithium-Ion Battery

NASA Aerospace Battery Workshop
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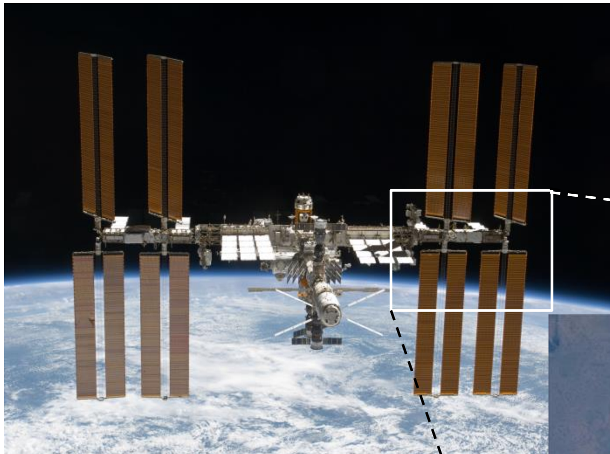
ISS Li-Ion Battery - Outline

- Configuration of Existing ISS Electric Power System
- Timeline of Li-Ion Battery Development
- Battery Design Drivers
- Technical Definition Studies
- Cell Selection
- Safety Features
- Final Flight Adapter Plate and Battery Design
- Battery Charge Control and LEO Cycle Test Data
- Cell and ORU Life Test
- Current Status

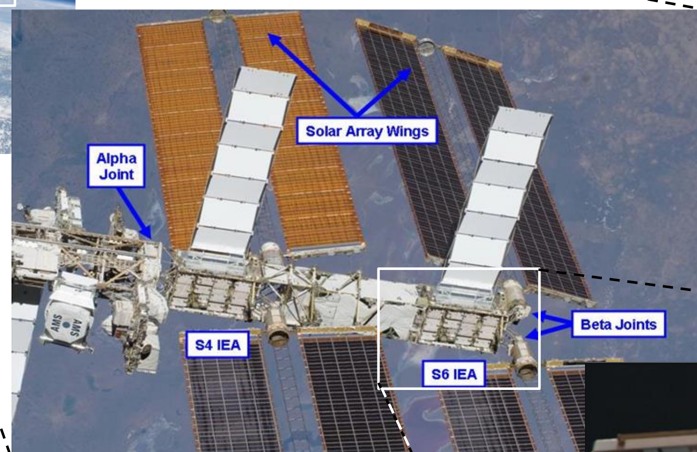
ISS Configuration - Battery Locations

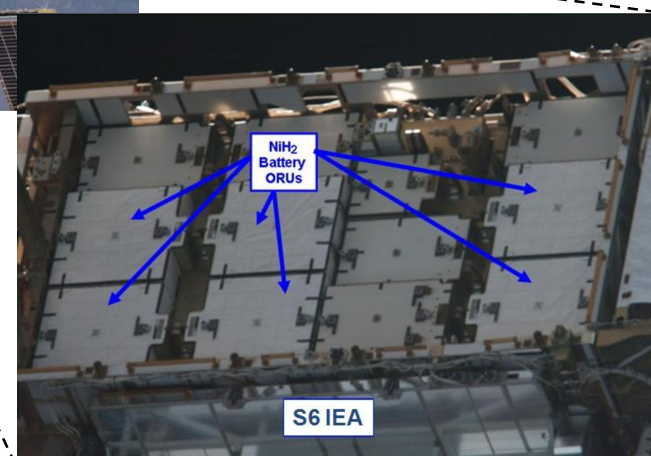


Batteries are located in the 4 Integrated Equipment Assemblies (IEAs)

2 Power Channels per IEA

8 Power Channels total



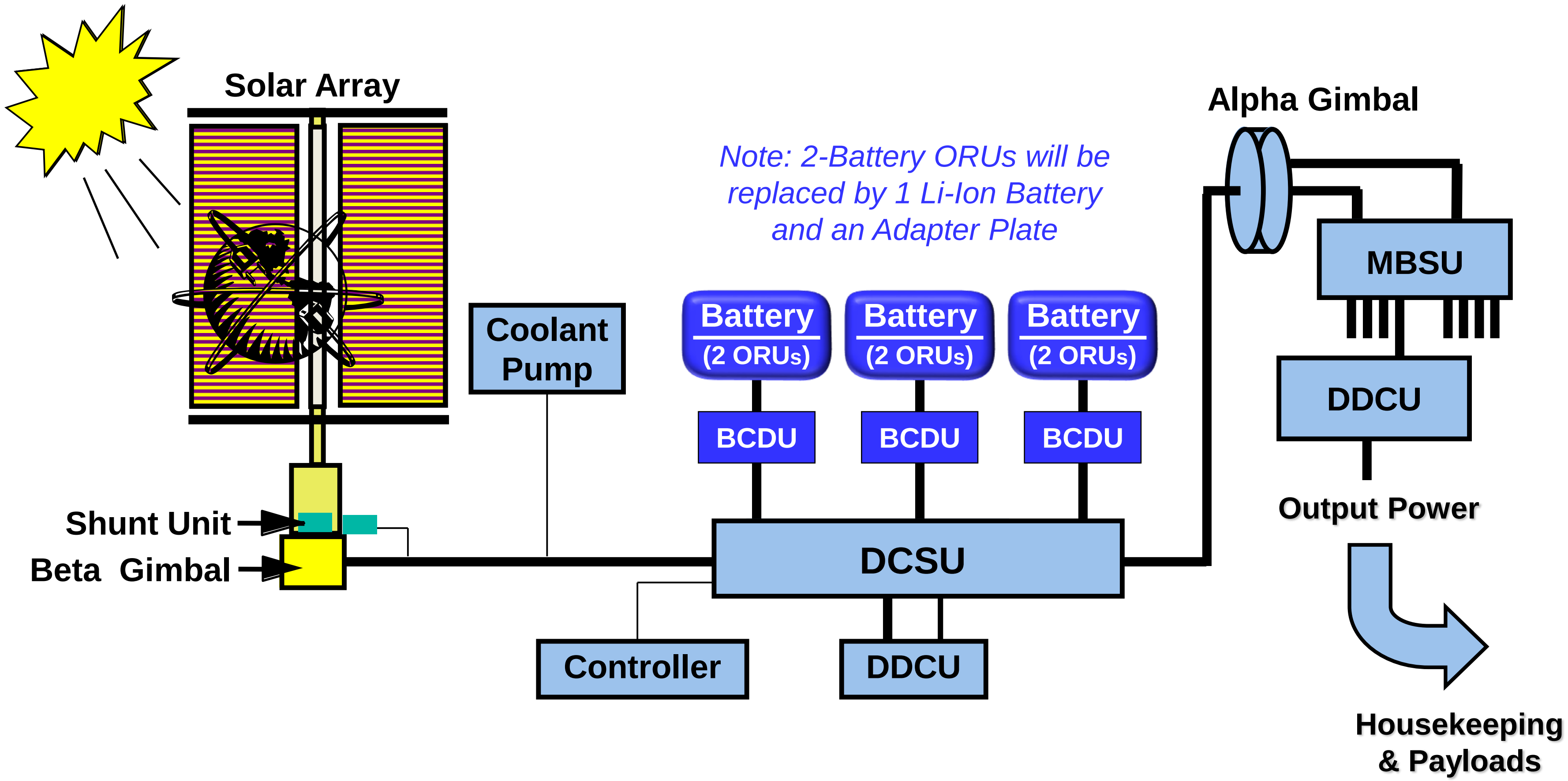


6 $Ni\text{-}H_2$ ORUs per channel – 48 total

1 Li-Ion and 1 Adapter Plate to replace 2 $Ni\text{-}H_2$ – 24 total Li-Ion batteries

ISS Configuration - EPS Schematic

Electrical Power Channel – 1 of 8



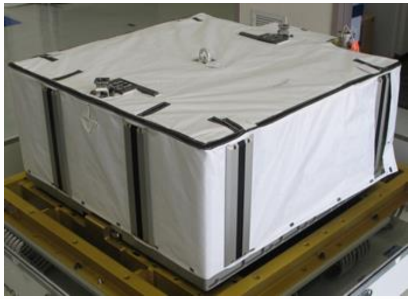
EPS:: Electric Power System
BCDU: Battery Charge / Discharge Unit
DCSU: DC Switching Unit
DDCU: DC-to-DC Converter Unit
MBSU: Main Bus Switching Units

ISS Li-Ion Battery Project Overview

- Battery ORU (Orbital Replacement Unit)
  - Battery ORU Design and Manufacture
  - Baseplate Design and Manufacture
  - Enclosure Design (HOU) and Manufacture (AASC)
  - Li-Ion Battery Cells (GS Yuasa)
  - Charge Control Electronics Design and Manufacture
- On-Orbit Adapter Plate (Atec)
- Flight Support Equipment (FSE) Interface Hardware
- Li-Ion Battery Status/Charging Unit (SCU)
- Software Updates (PVCA, PCS, PMCA, and CCS)
- Testing
  - ORU Verification and Qualification Testing
  - Battery Cell Qualification and Acceptance Testing
  - Battery ORU Life Testing
  - ISS Systems Integration Testing
  - Battery Cell Safety Characterization/Abuse Testing and Battery Cell Life Testing
  - Post Delivery ORU Freezer/Refrigerator Storage
- Automated Test Equipment Design and Manufacture

Color Key (Scope):

- ❑ *NASA*
- ❑ *Boeing*
- ❑ *ORU Supplier AR (Aerojet Rocketdyne)*
- ❑ *Joint Boeing/AR*

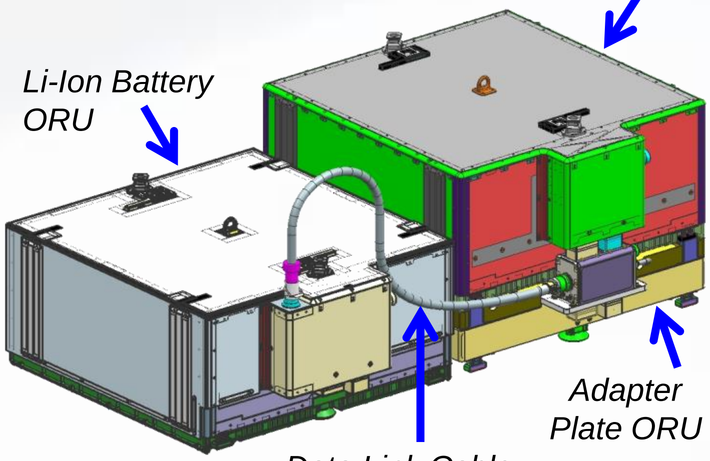


Timeline of ISS Li-Ion Development

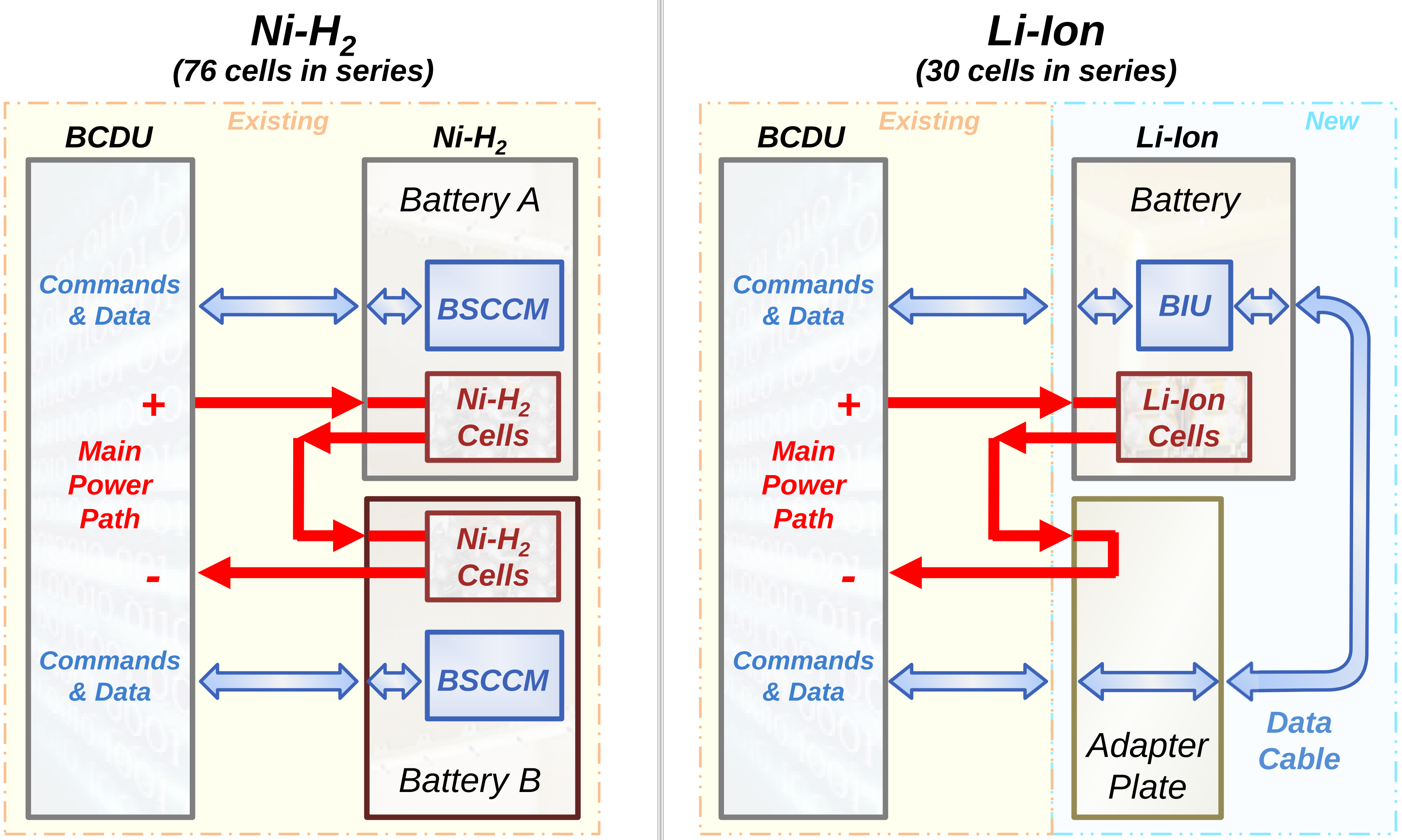
- **2009-2010 –** Preliminary risk and feasibility studies
- **December 2011 -** ISS Program Authority To Proceed with design, development and the fabrication of 27 Li- Ion ORUs and 25 on-orbit Adapter Plate ORUs
- **Jan-Jun 2012** - Cell Safety Testing and Cell Qualification
- **July 2012** - Final cell down-select
- **December 2012** - System Preliminary Design Review
- **November 2013** - System Critical Design Review
- **March 2016** - First flight Li-Ion battery delivered to Kennedy Space Center for shipment to Tanegashima, Japan

ISS Li-Ion Battery Key Design Drivers

- 1 Li-Ion battery ORUs replaces 2 Ni-H_2 ORUs
  - Li-Ion ~15 kWh vs. Ni-H2 ~4 kWh each
- Launch on Japanese HTV
- 6 year battery storage life requirement
- 10 year/60,000 cycle life target (minimum 48 A-hr capacity at end of life)
  - ORU will have cell balancing circuitry
  - ORU will have adjustable End of Charge Voltage (EOCV)
- Maximum battery ORU weight ~430 lbs
- Non-operating temperature range (Launch to Activation): -40 to +60 °C
- No changes to existing IEA interfaces and hardware
  - Use existing mounting, attachment, electrical & data connectors
  - Use existing Charge/Discharge Units and Thermal control systems

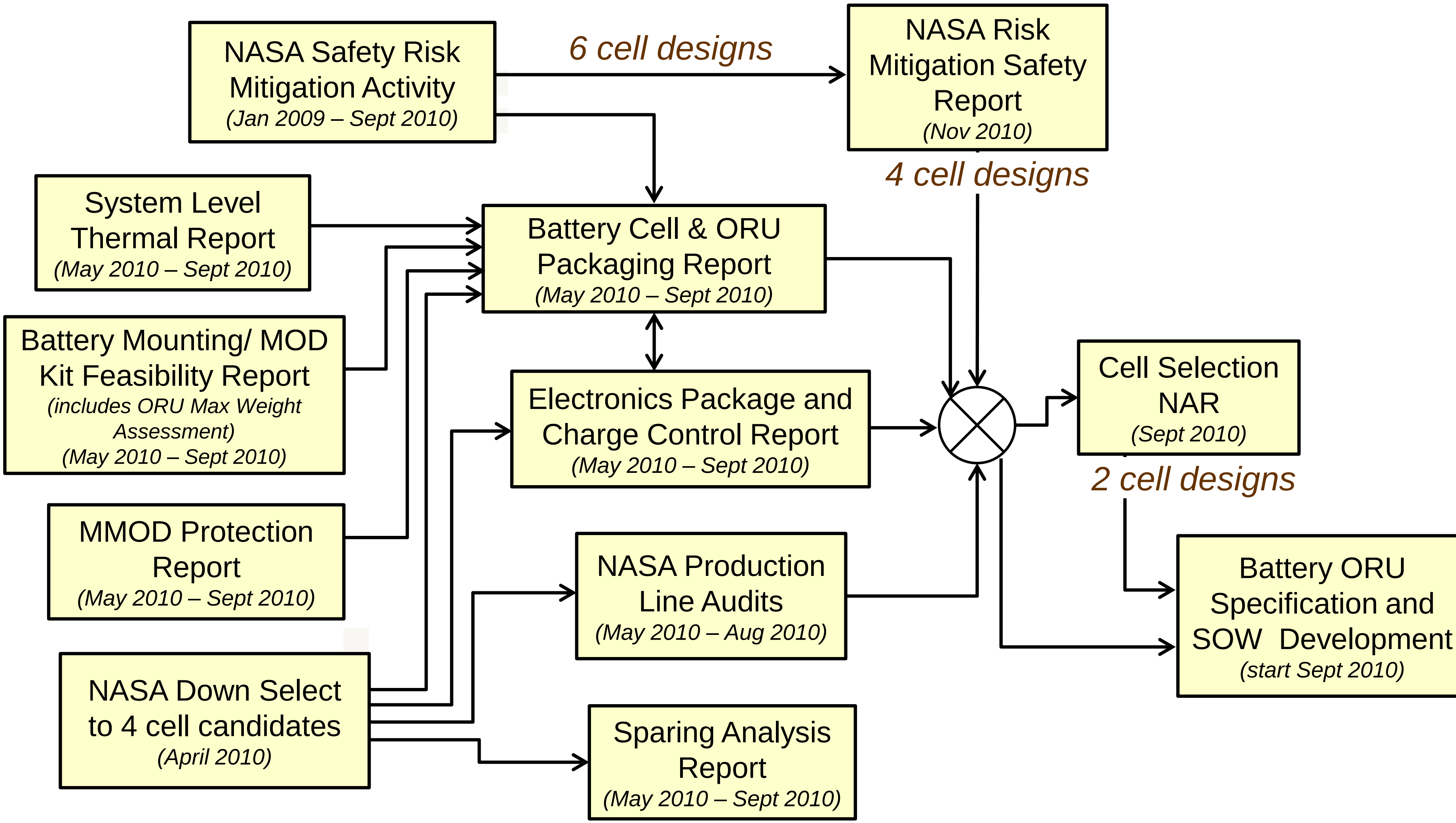


ISS Upgrade to Li-Ion



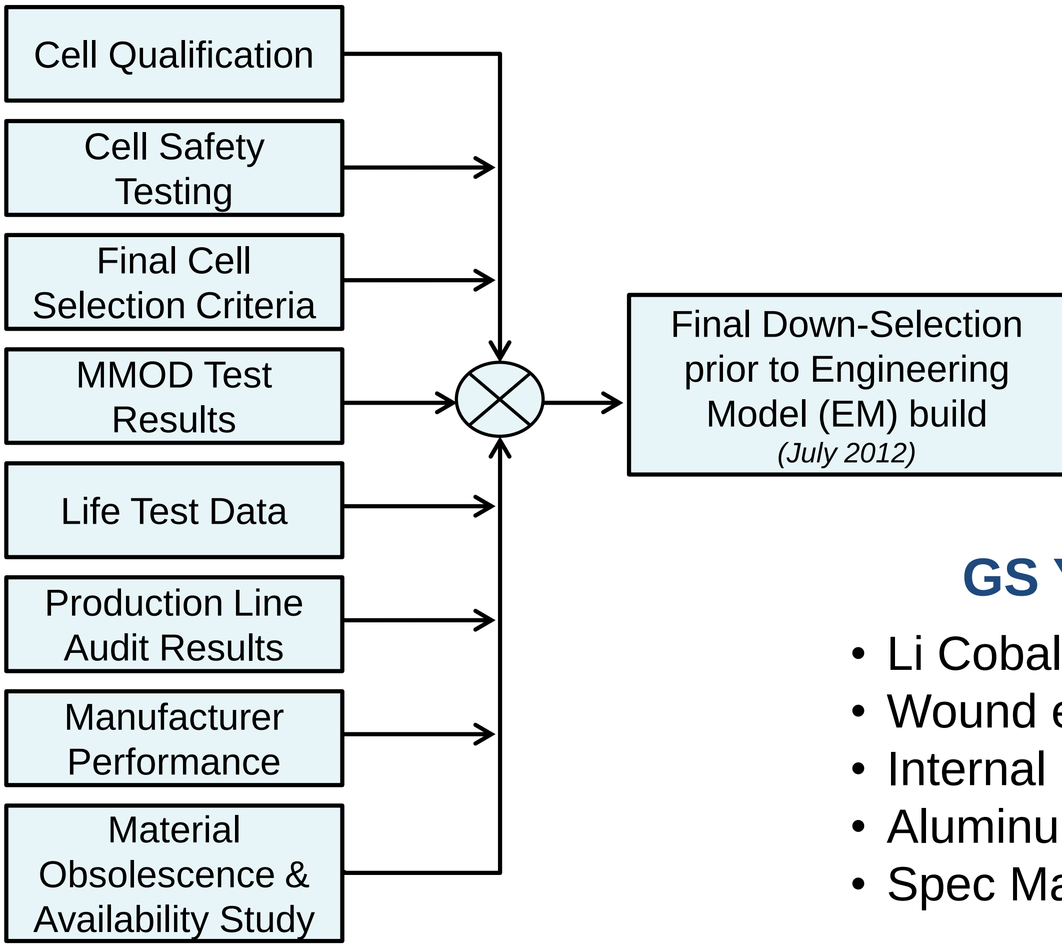
BCDU: Battery Charge / Discharge Unit
BIU: Battery Interface Unit
BSCCM: Battery Signal Conditioning and Control Module

ISS Li-Ion Technical Definition Studies



ISS Li-Ion Cell Final Down-Select

- Two designs taken through qualification, with down-selection made prior to EM build



GS Yuasa 134 A-hr cells

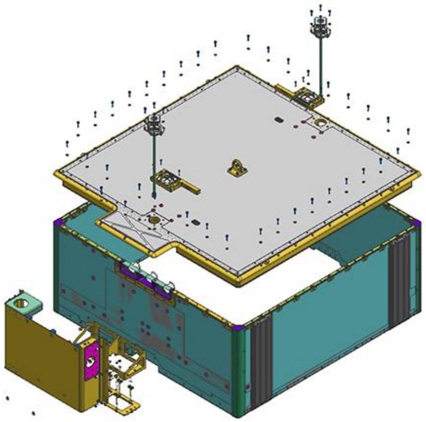
- Li Cobalt Oxide / Carbon Graphite
- Wound elliptical prismatic electrode
- Internal Fusible link
- Aluminum Case, 50 x 130 x 263 mm
- Spec Mass: 3530 grams (~7.8 lb)

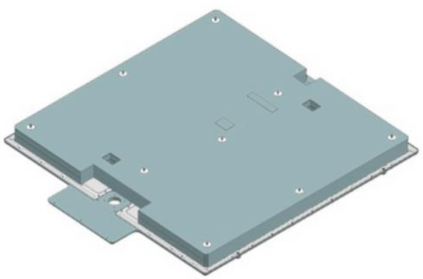
ISS Li-Ion Battery Safety Features

Battery-Level Safety Features

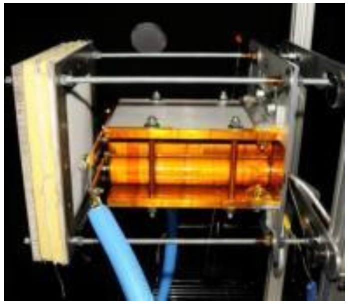
- 2 independent controls vs. thermal runaway (2 fault tolerant)
- Voltage and temperature monitoring of all 30 cells
- Circuit protection/fault isolation at the individual cell level for both high/low voltage and high temperature
- Physical separation between cell pairs and 10 packs
  - Thermal radiant barriers between cell pairs
- Controlled direction of cell vents - prevent damage to cold plate, adjacent cells and IEA hardware
  - ORU pressure relief/flame trap to prevent ORU over-pressurization but contain flame in the event of a cell vent
- MMOD shielding in ORU and empty ORU slot
- Dead face device to remove power from output connector during ground or EVA handling
- Non propagation of failures beyond Battery ORU

Safety Features - MMOD Shielding

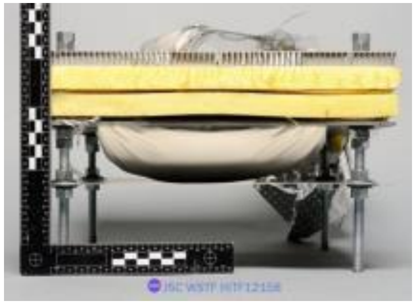




MMOD Shield

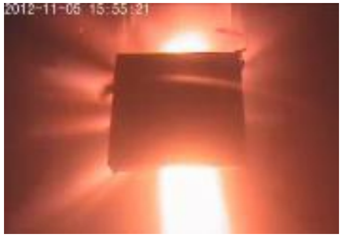


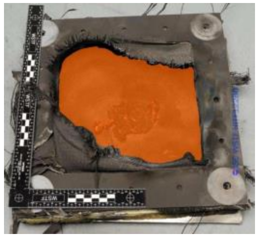
MMOD test setup



Ballistic Limit Testing

Over Match - Penetration testing
10 mm 2017-T4 Aluminum Sphere @ 6.86 km/s

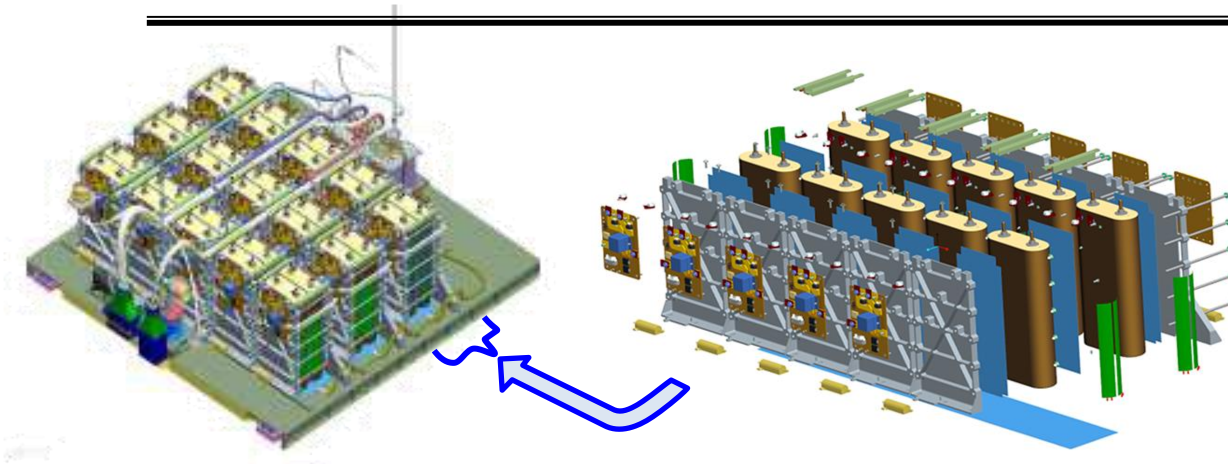




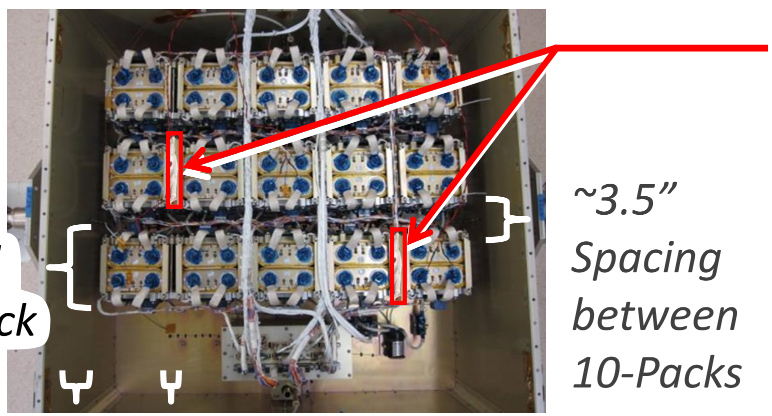
Overcharge Containment Testing

Note: Existing Ni-H_2 does not have MMOD protection

Safety Features - Radiant Heat Barriers



- ORU Layout – 3 Cell "10-Packs" and 12 Radiant Barriers



Radiant Heat Barrier (12 per ORU)

- Higher margin against thermal runaway propagation
- 1 barrier between each cell pair
- Reflects 787 reach-back safety additions

ISS Li-Ion Cell Safety Features

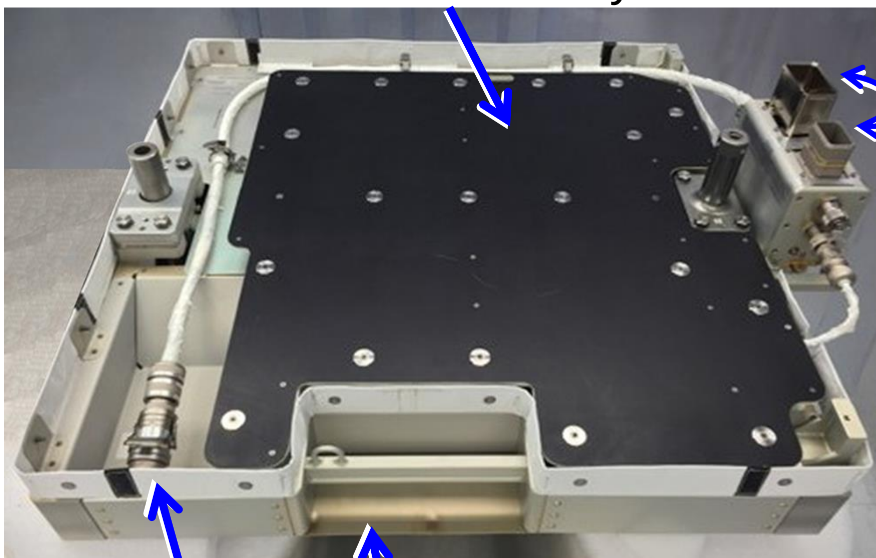
Cell-Level Safety Features and Controls

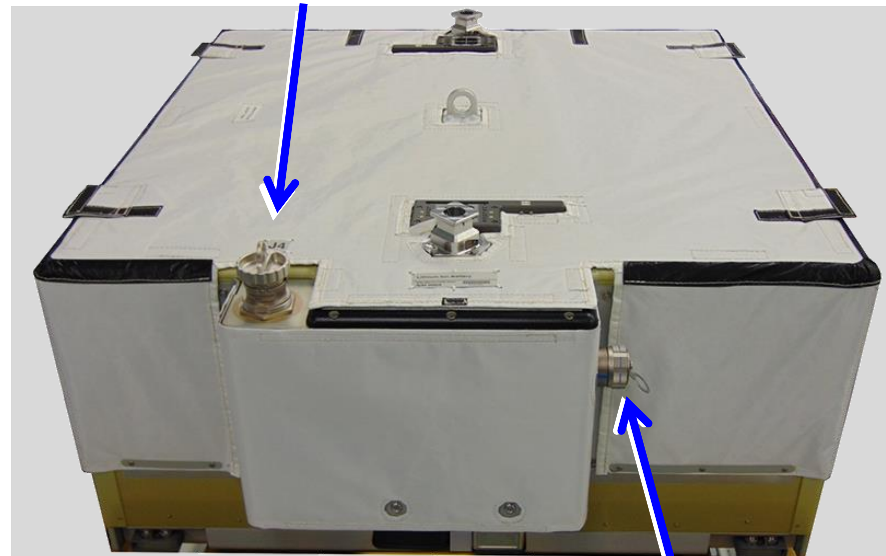
- Manufacturing Process controls include 100% materials screening and chemical analysis plus annual configuration/production line audits
- 100% cell acceptance testing
  - Cell Matching performed based on ATP characteristics
- 2% of cells in each lot in simulated LEO life cycle testing
- 1% of cells in each lot undergo 100, 100% DOD cycles, followed by DPA
- Cell vent before burst and directional vent away from base plate and adjacent cells
- Individual cell fusing (internal fusible link)
- Shutdown separators between electrode windings
- Case neutral and electrically insulated from ORU structure

ISS Li-Ion Orbital Replacement Units

Heater Matt
Heater Plate Assembly

J4
Connector





P4 Connector
(stowed for launch)

EVA
Hand Hold

P1 & P2
Connectors

J3 Test
Connector

Adapter Plate ORU

Dimensions (LxWxH):	~ 41” x 36” x 15”
Spec Weight:	85 Lbs

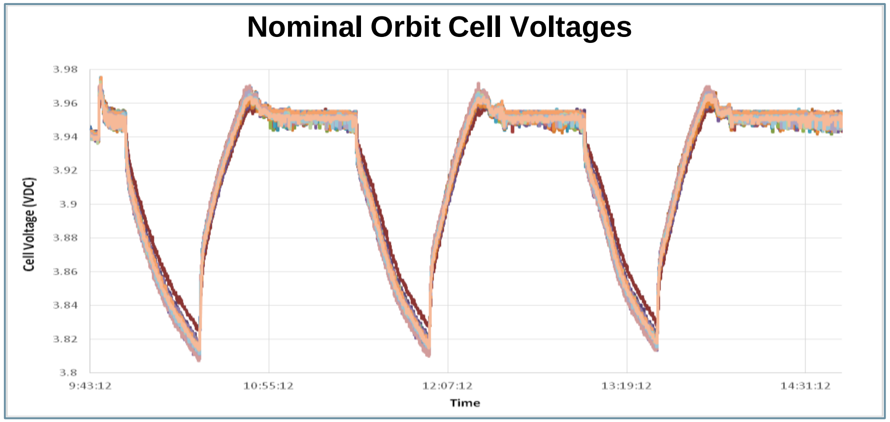
Li-ion Battery ORU

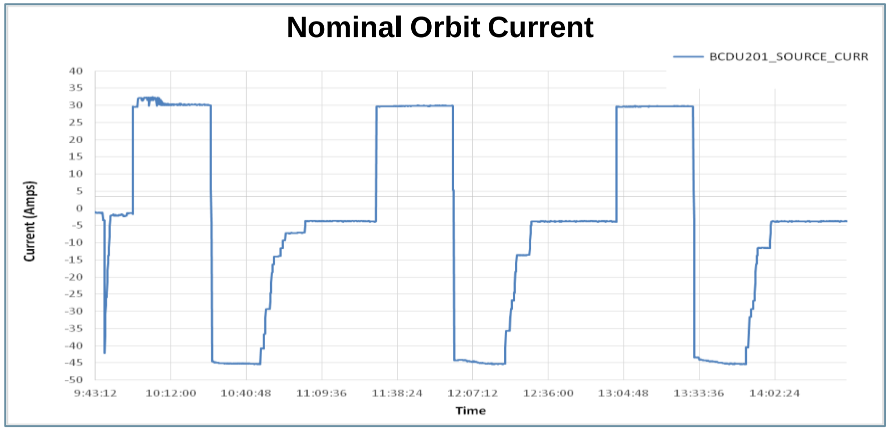
Dimensions (LxWxH):	~ 41” x 37” x 21”
Spec Weight:	435 Lbs

ISS Li-Ion Charge Control and Cycling

- Li-Ion charge current profile based on cell voltages
- Cell bypass/balancing at EOCV every orbit
- EOCV is ground command-able

Charge Current Profile		
	Highest of the Cell Terminal Voltages	**Charge Current**
Point 1	EOCV + 19mV	55
Point 2	EOCV + 19mV	49
Point 3	EOCV + 18mV	44
Point 4	EOCV + 17mV	39
Point 5	EOCV + 16mV	36
Point 6	EOCV + 15mV	33
Point 7	EOCV + 14mV	30
Point 8	EOCV + 13mV	26
Point 9	EOCV + 12mV	22
Point 10	EOCV + 11mV	19
Point 11	EOCV + 10mV	16
Point 12	EOCV + 9mV	13
Point 13	EOCV + 8mV	10
Point 14	EOCV + 7mV	7
Point 15	EOCV + 6mV	4
Point 16	not applicable	1

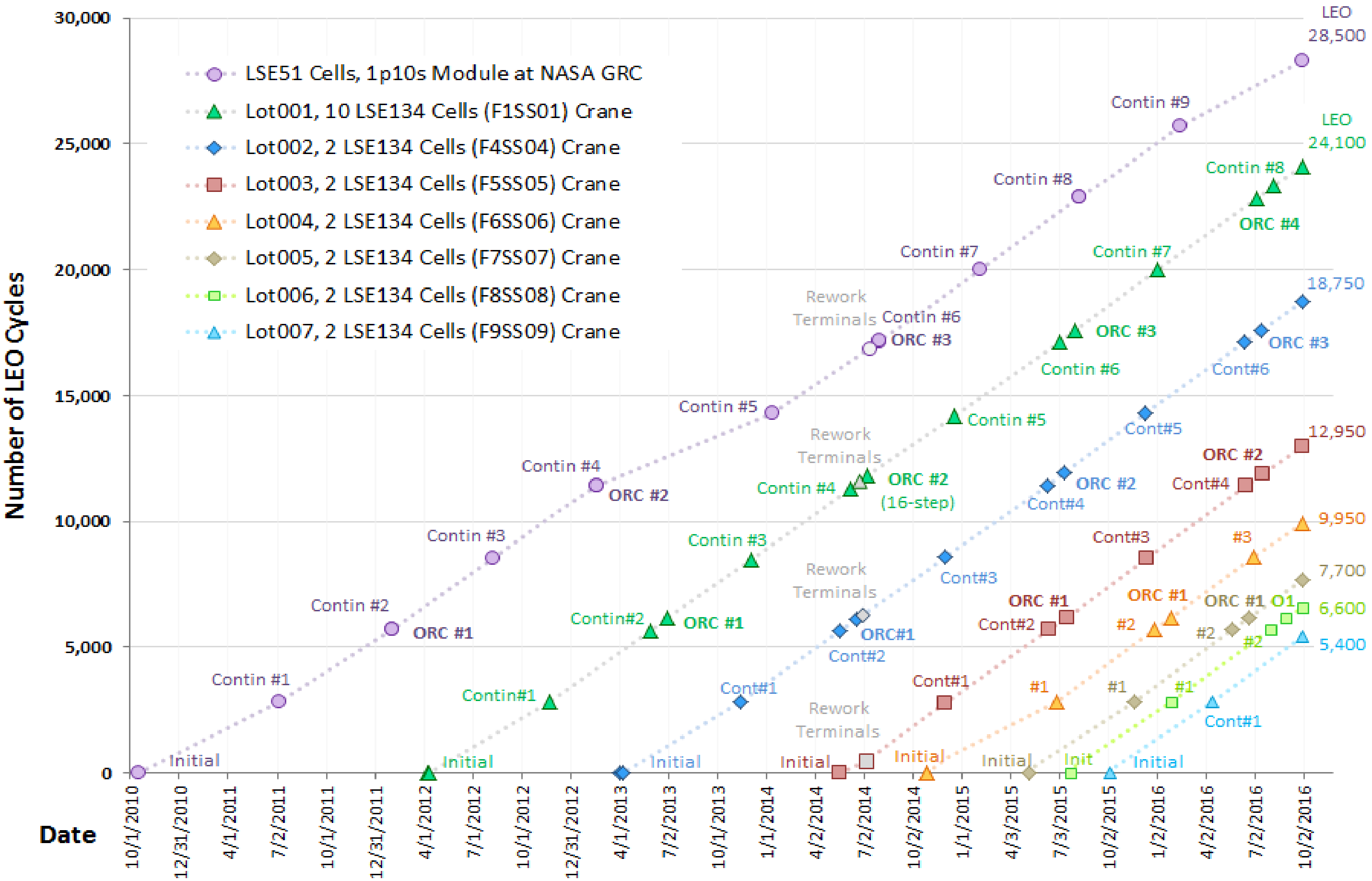




Life Test Program

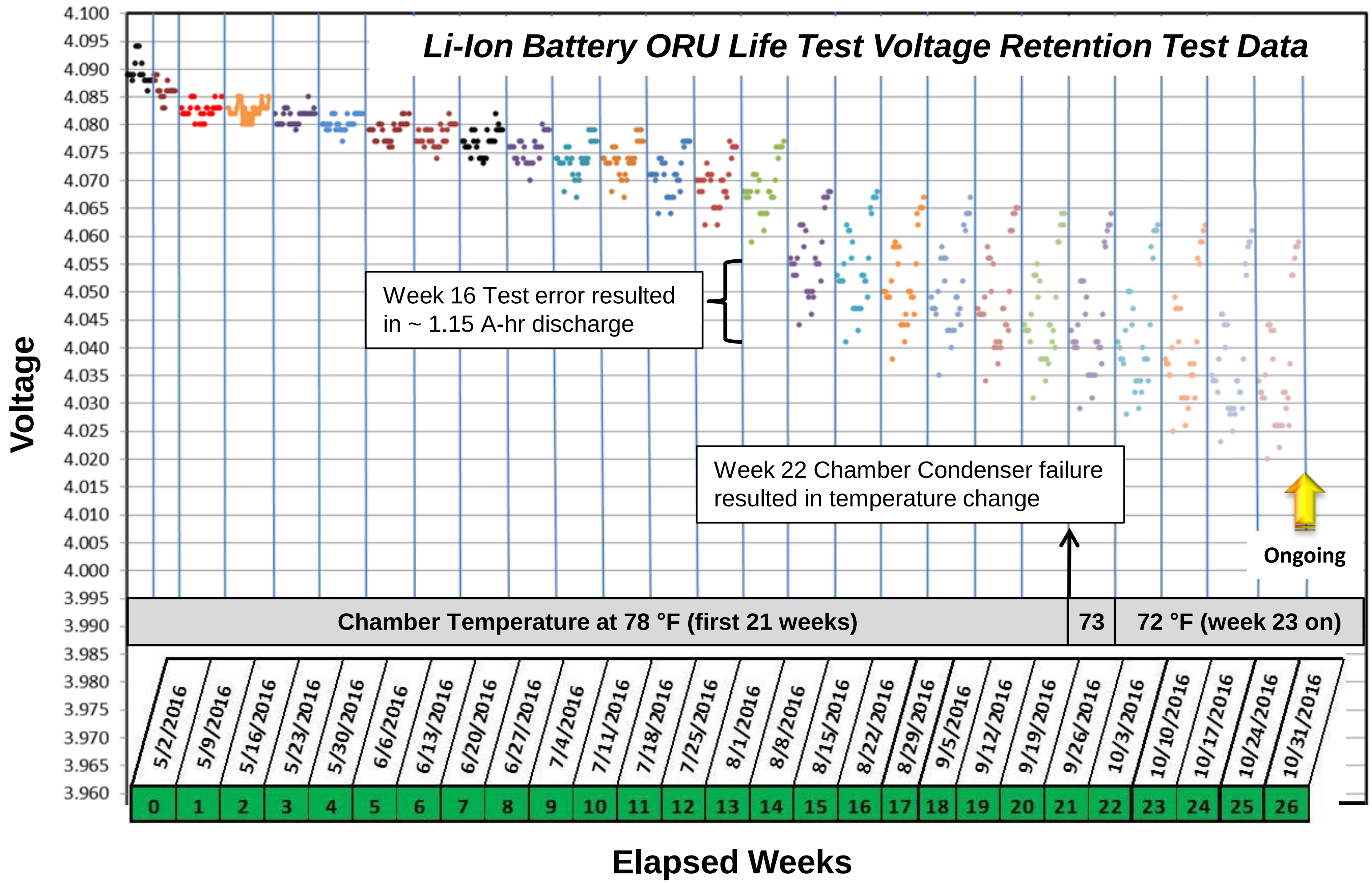
- Cell Life Testing performed at Crane and at GRC

Cell Testing at NSWC Crane Lab and NASA-GRC

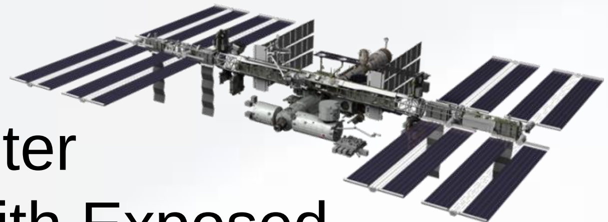


Life Test Program

- ORU Life Testing at Aerojet Rocketdyne



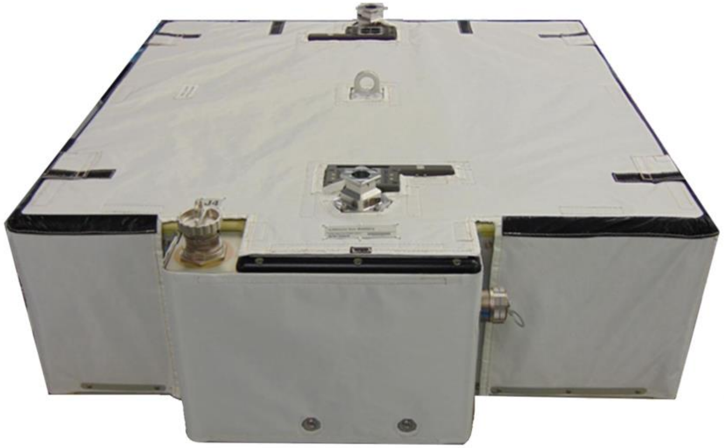
ISS Li-Ion Flight Battery Status



- 6 Flight Li-Ion Adapter Plates integrated with Exposed Pallet in Japan, Tomioka: April 2016
- 6 Flight Li-Ion Batteries integrated with Exposed Pallet in Japan, Tanegashima: May 2016
- Final charge to 4.1V: May-June 2016
- Launch on HTV: NET December 2016
  - Each IEA will have 3 Li-Ion ORUs and 3 Adapter Plate ORUs
- Installation and start-up on ISS: NET Dec. 2016 – Jan. 2017

ISS Li-Ion Battery Future Plans

- Data analysis for NESC (NASA Engineering & Safety Center) Thermal runaway propagation test performed October 2016 at the White Sands Test Facility
- Launch of six Li-Ion Batteries and six Adapter Plates in 2017, 2018, 2019 to provide a full complement on ISS



➢ *Ready for successful and safe operation*

In Closing

- Acknowledgements
- Questions?