

# CONFIDENCE LEVEL BASED APPROACH TO TOTAL DOSE SPECIFICATION FOR SPACECRAFT ELECTRONICS

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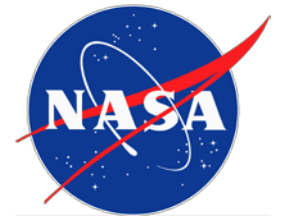
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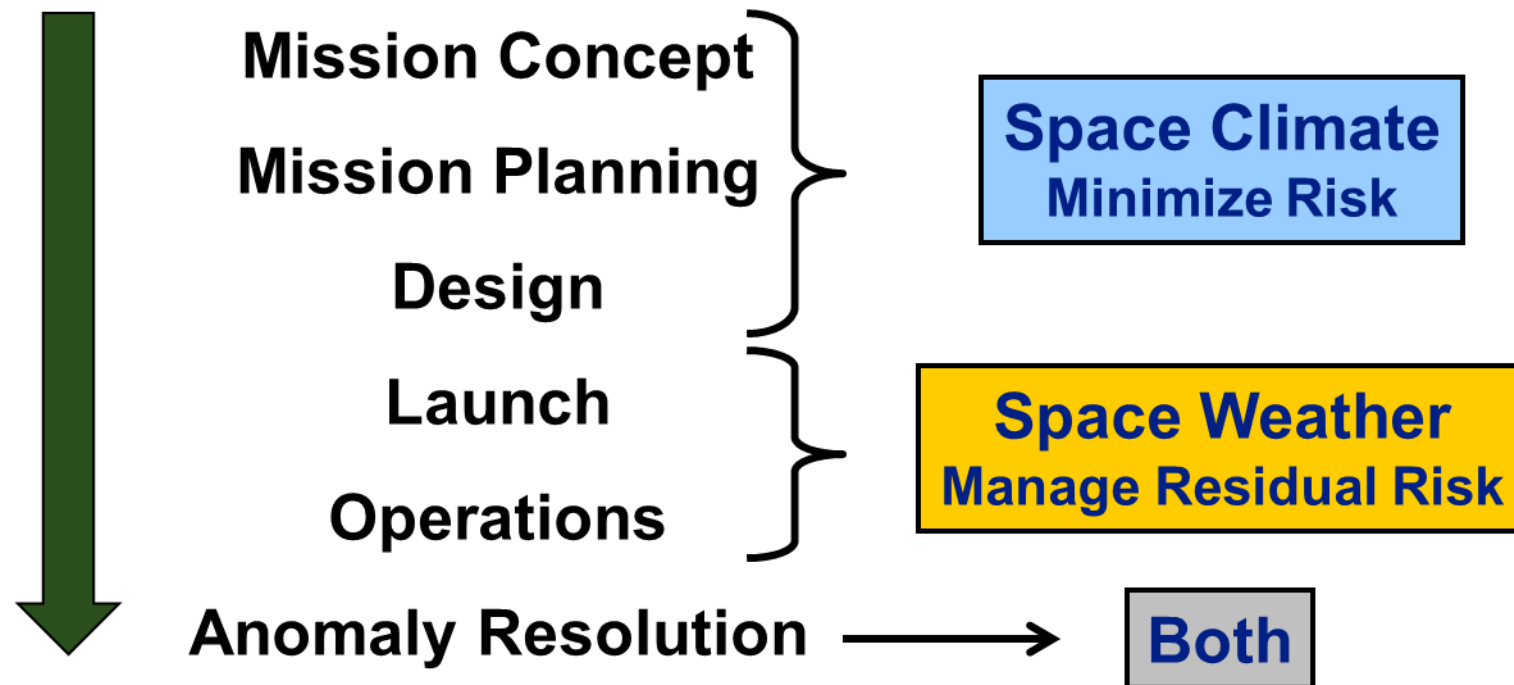
Supported by the NASA Living With a Star Space Environment Testbed Program



# Outline

- **Background**
- **Device Failure Distributions in Total Dose**
- **Total Dose Distributions in Space**
- **Device Failure Probability during a Mission**
- **Conclusions**
  - **Failure Probability ( $P_{fail}$ ) vs. Radiation Design Margin (RDM)**

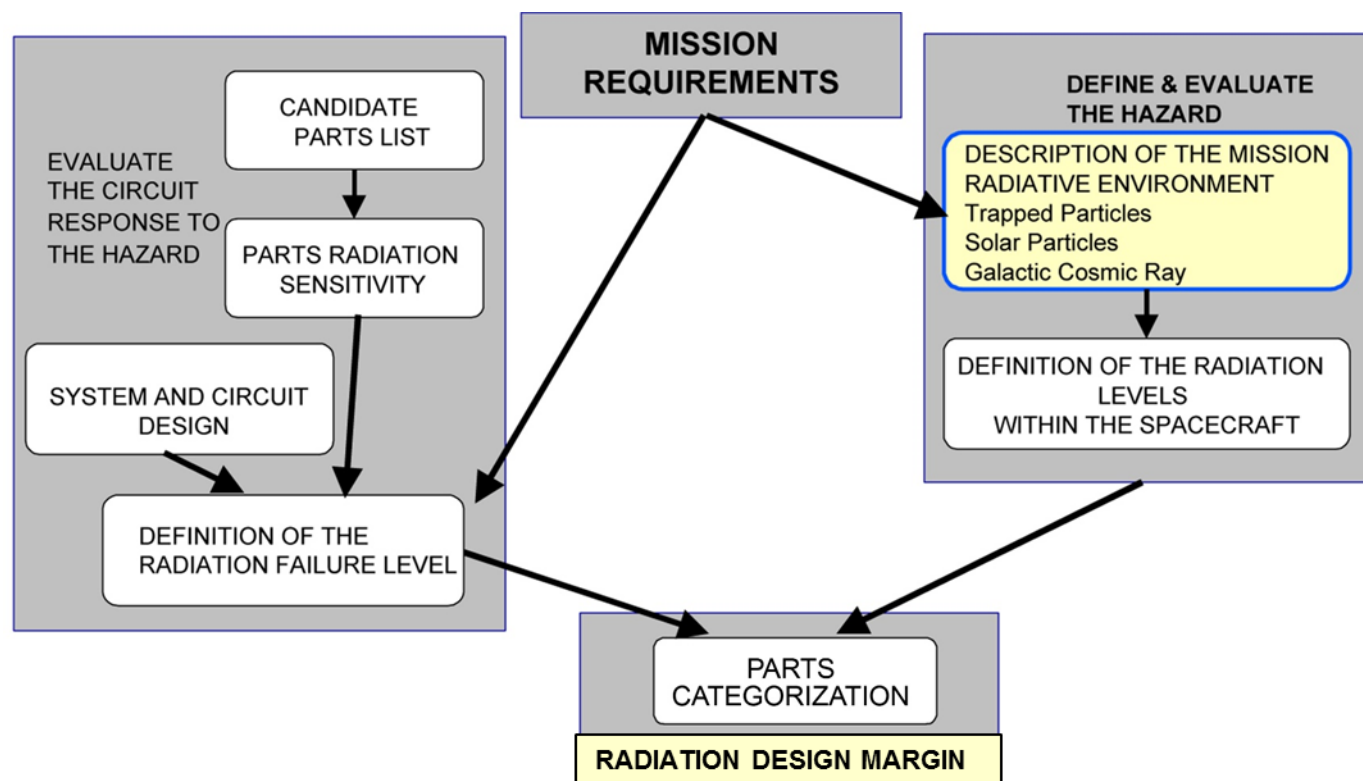
# Space Environment Model Use in Spacecraft Life Cycle

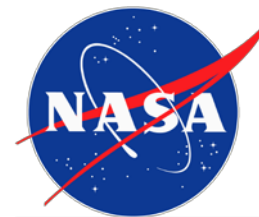




# Radiation Hardness Assurance Overview

- **Starting with mission requirements, methodology consists of 2 branches of analyses that lead to parts categorization**
  - **Parts analysis**
  - **Environment analysis**





# Radiation Hardness Assurance Overview

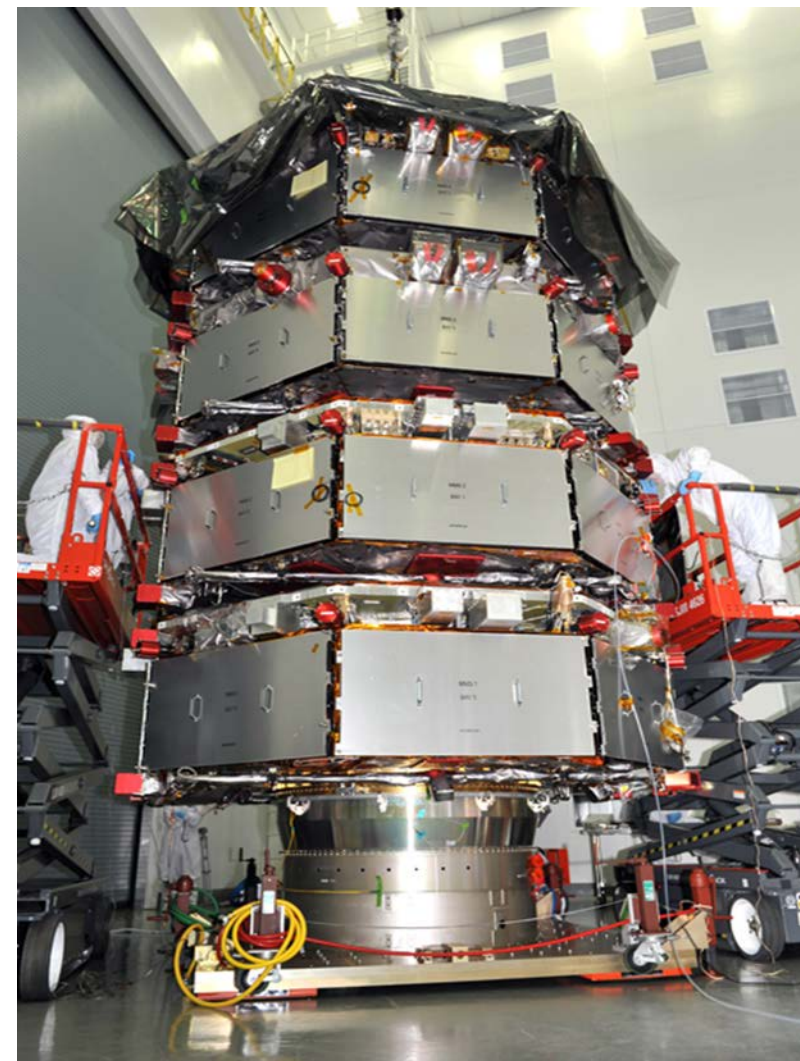
- **Parts are categorized for flight acceptability and possible radiation lot acceptance testing by Radiation Design Margin (RDM).**
- **$RDM = R_{mf} / R_{spec}$**
- **$R_{mf}$  is mean failure level of part**
- **$R_{spec}$  is total dose level of space environment**
- **Difficulties can arise because**
  - **Part failure levels can vary substantially from the mean, especially COTS**
  - **Environment is dynamic and must be predicted years in advance**
- **RDM based approach results from use of deterministic AP8/AE8 trapped particle models**
- **RDM used as a “catch-all” to cover all uncertainties in environment and device variations**
- **Propose modified approach**
  - **Use device failure probability during a mission instead of RDM**



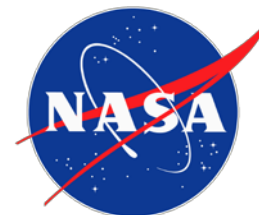
## 4 stacked MMS spacecraft

# Devices Tested

- **Solid State Devices, Inc.**  
**SFT2907A bipolar transistors**
  - Used for high speed, low power applications
  - 10 devices TID tested for MMS project at NASA/GSFC gamma ray facility to 100 krad(Si)
- **Amptek, Inc.** HV801 optocouplers
  - GaAlAs parts manufactured in liquid phase epitaxially grown process
  - 6 devices DDD tested for JUNO project at UC Davis Cyclotron with 50 MeV protons

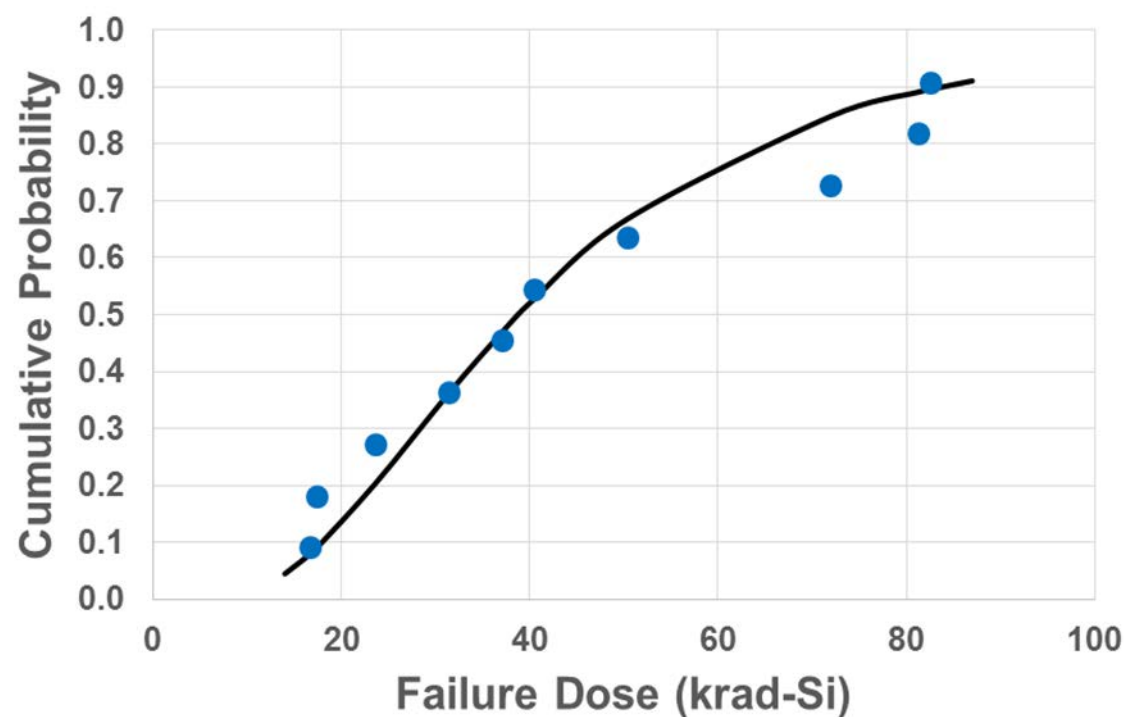
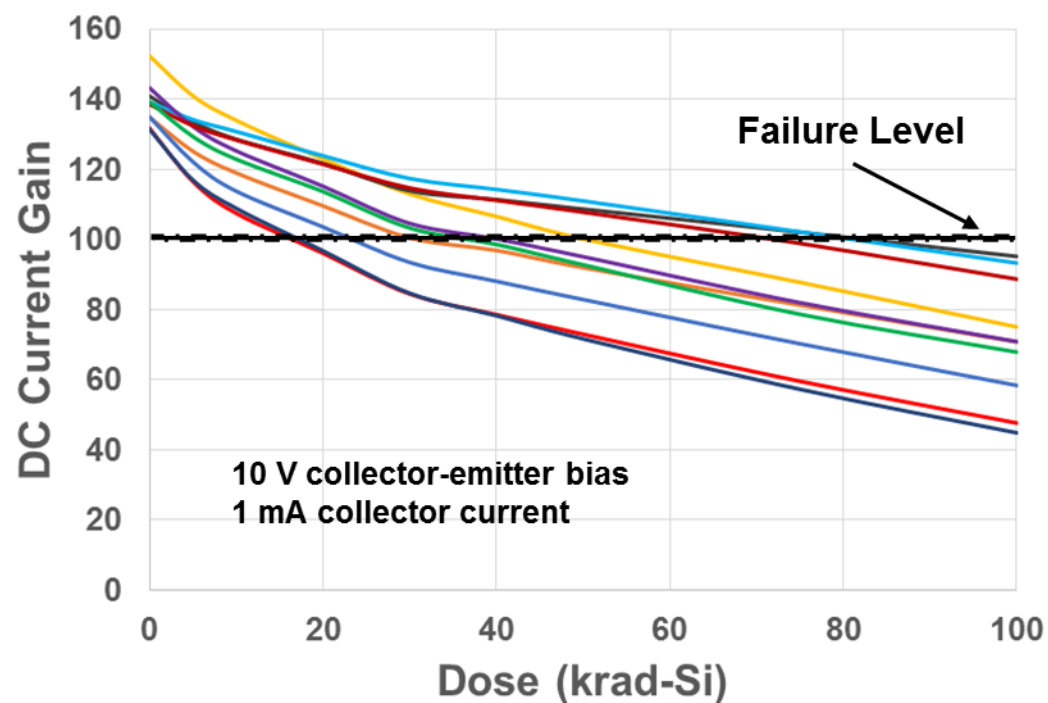


Credit: <http://mms.gsfc.nasa.gov>



# Device Failure Distribution

## SFT2907A Bipolar Transistors

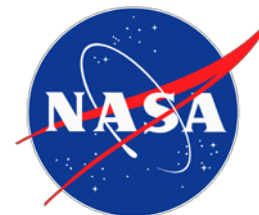




# Total Dose Probability Distribution Calculations

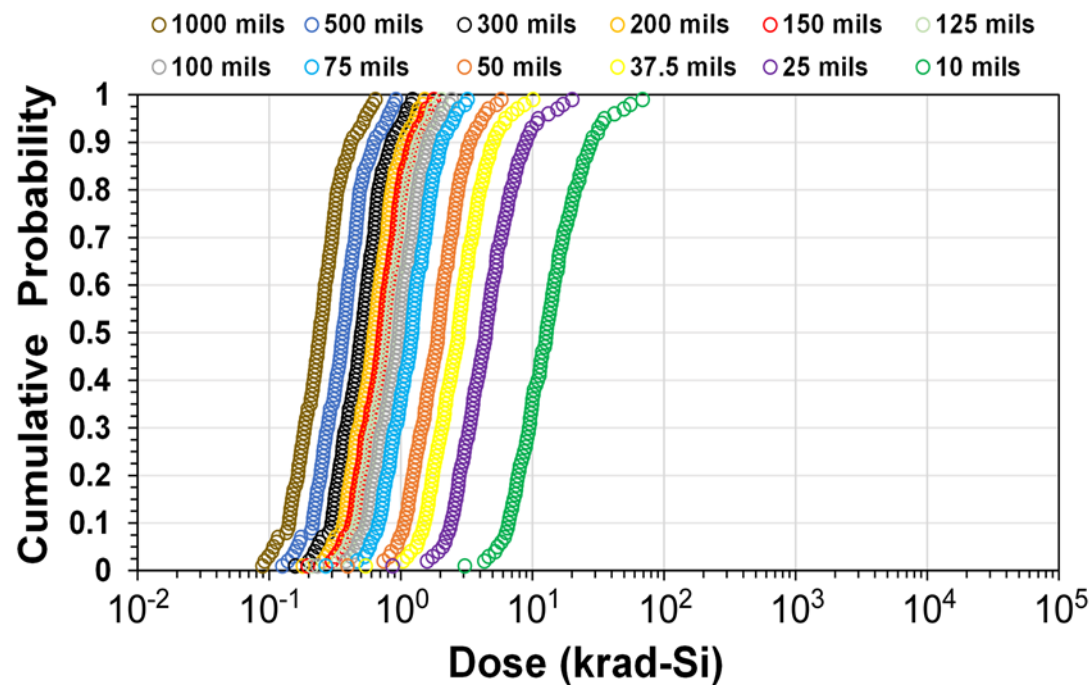
- **TID and DDD probability distributions were calculated for each orbit and mission duration for confidence levels ranging from 1 to 99%**
  - **AP9/AE9 Monte Carlo code used to simulate 99 histories for each case**
  - **ESP solar proton calculations done for 1 to 99% confidence levels**
  - **All energy spectra were transported through shielding levels from 10 to 1000 mils Al using NOVICE code and converted to doses**
  - **TID and DDD for each radiation were separately ranked for confidence levels ranging from 1 to 99% and summed for same confidence and shielding levels**



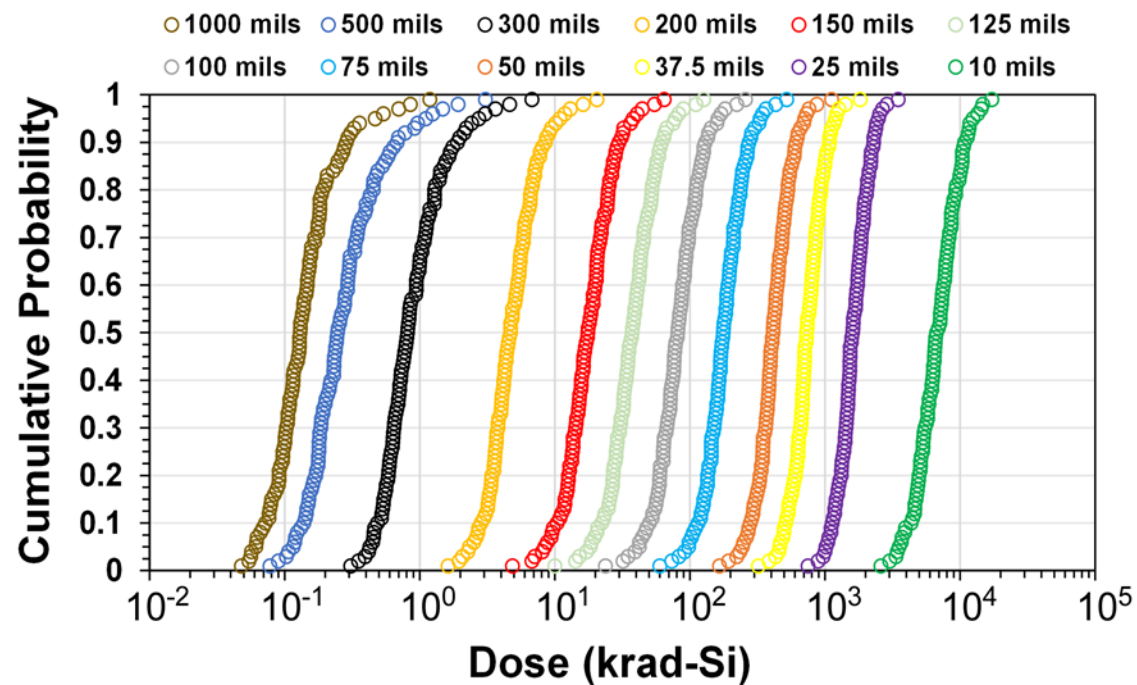


# TID Probability Distributions for 1 Year 10 – 1000 mils Aluminum

## Low Inclination LEO



## GEO





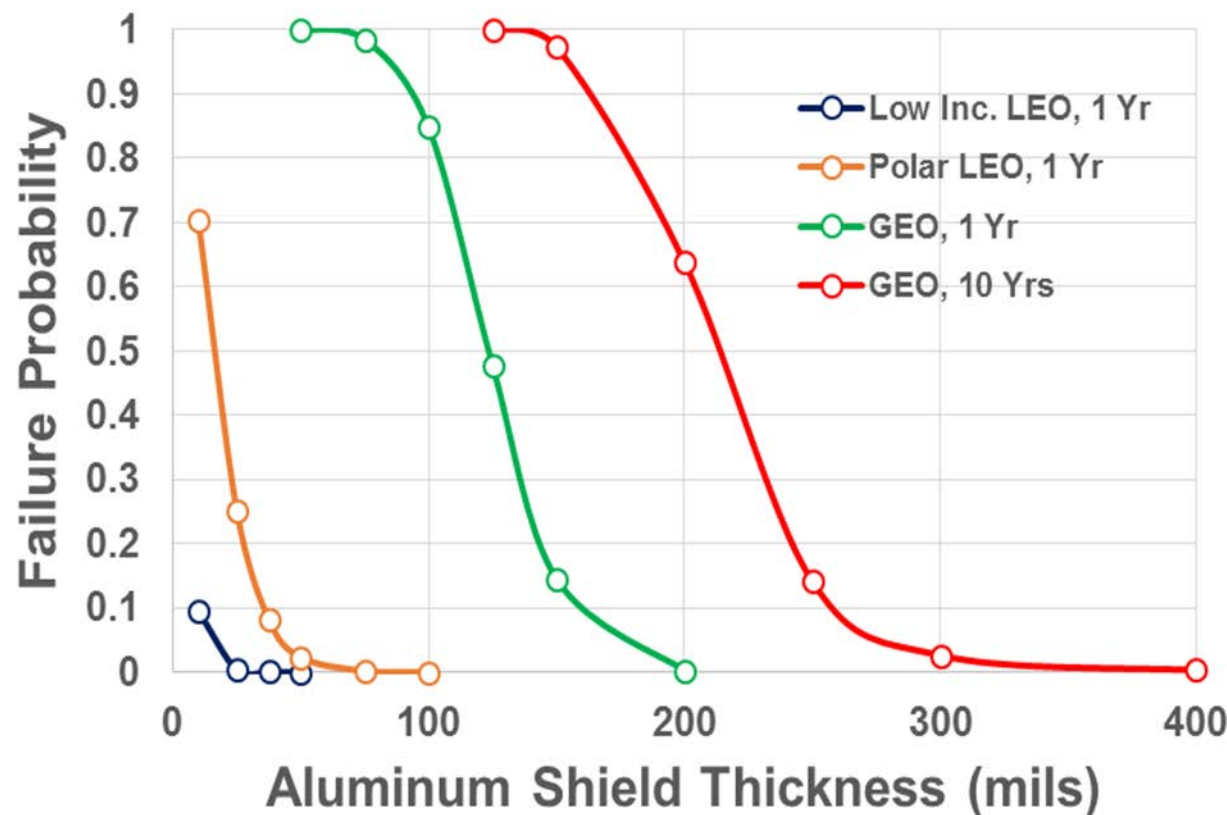
# Failure Probabilities SFT2907A Bipolar Transistor

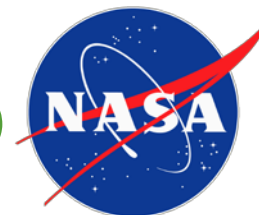
$$P_{\text{fail}} = \int [1 - H(x)] \cdot g(x) dx$$

$H(x)$  = CDF for environment dose

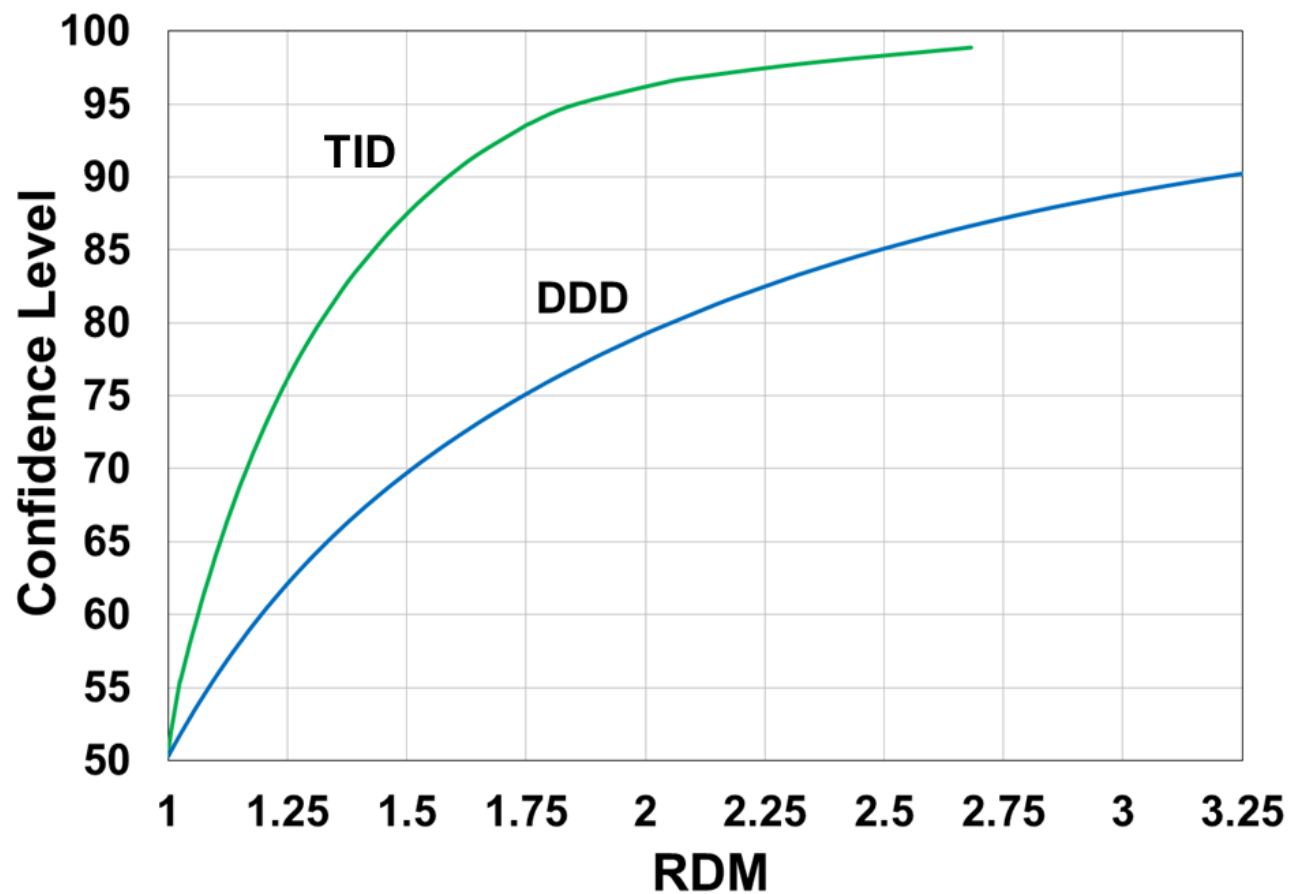
$g(x)$  = PDF for device failure

Failure probability ( $P_{\text{fail}}$ ) is the probability of a total dose failure during a mission





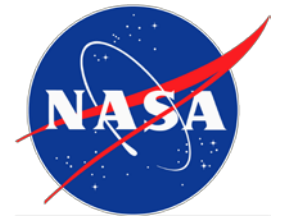
# Confidence Level vs. RDM for 10 years in GEO 200 mils Al shield





# Conclusions

- **An approach to total dose radiation hardness assurance was developed that includes variability of the space radiation environment.**
- **Examples showed radiation environment variability is at least as significant as variability of total dose failures in devices measured in the laboratory.**
  - **New approach is more complete**
  - **Uses consistent evaluation of each radiation in the space environment through use of confidence levels**
- **Advantages of using  $P_{fail}$  instead of RDM are:**
  - **$P_{fail}$  is an objectively determined parameter because complete probability distributions are used to calculate it; gives designers more trade space**
  - **Better characterization of device radiation performance**
  - **Allows direct comparison of the total dose threats for different devices and missions, regardless of whether degradation is due to TID or DDD**
  - **More amenable to circuit, system and spacecraft reliability analysis**

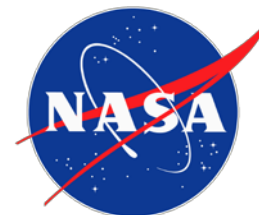


# Acronyms

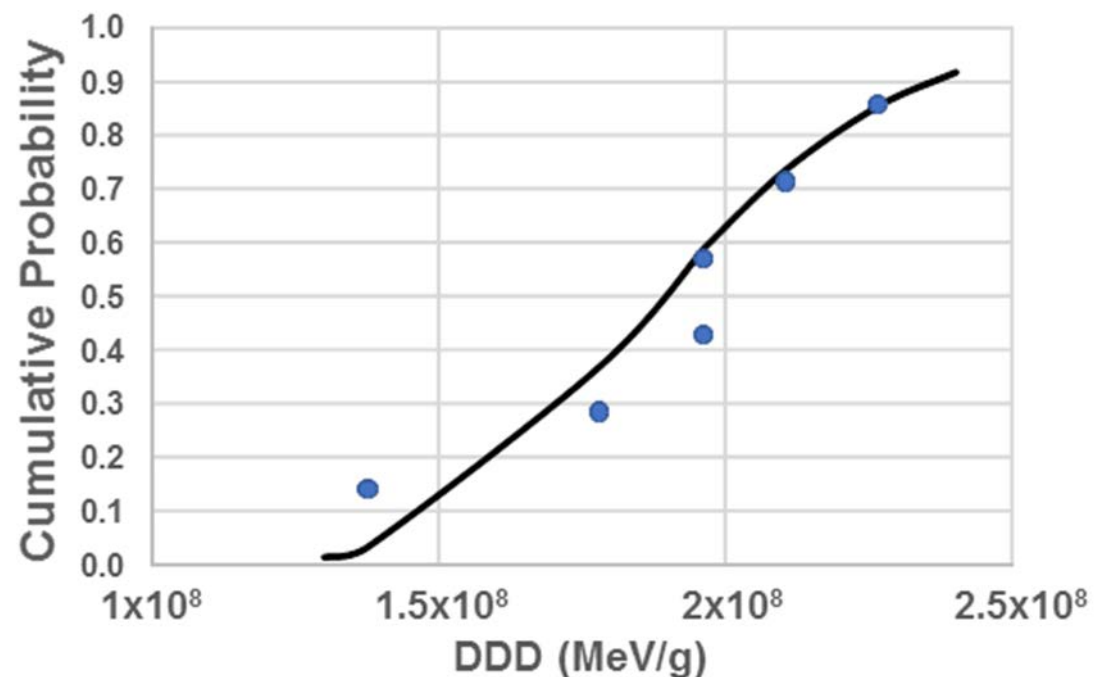
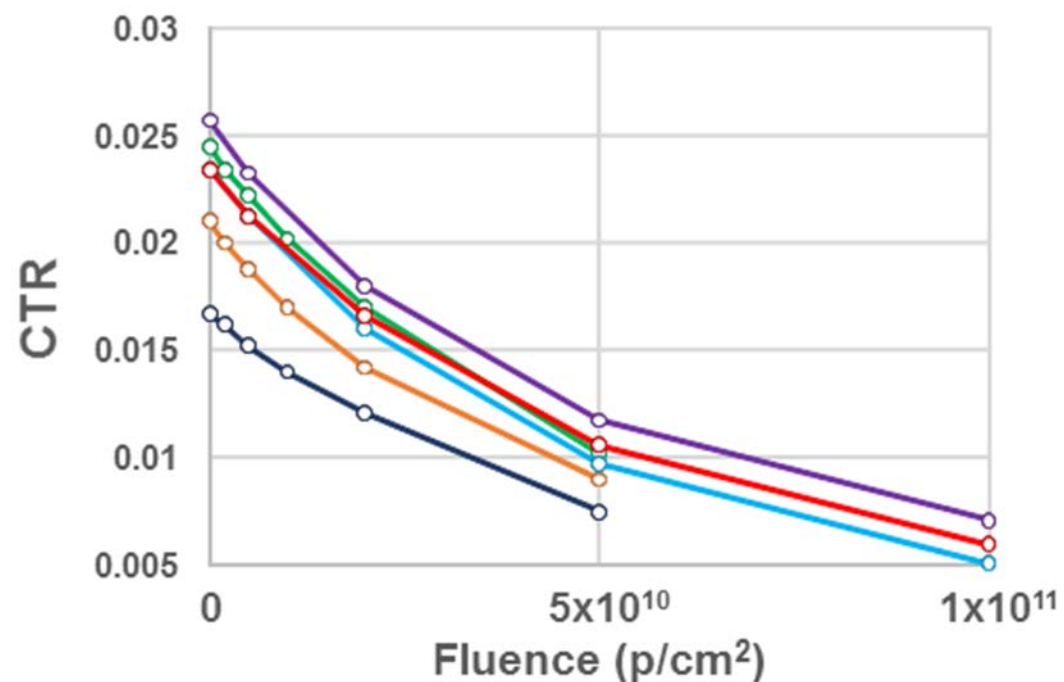
- **AE9 – Aerospace electron model-9**
- **AP9 – Aerospace proton model-9**
- **CDF – cumulative distribution function**
- **COTS - commercial off the shelf**
- **DDD – displacement damage dose**
- **ESP – Emission of Solar Protons (model)**
- **FP – failure probability**
- **GEO – geostationary Earth orbit**
- **HST – Hubble Space Telescope**
- **JUNO – JUpiter Near-polar Orbiter**
- **LEO – low Earth orbit**
- **MMS – Magnetospheric MultiScale**
- **NOVICE – Numerical Optimizations, Visualizations and Integrations on Computer Aided Design (CAD)/Constructive Solid Geometry (CSG) Edifices**
- **PDF – probability density function**
- **RDM – radiation design margin**
- **TID – total ionizing dose**

# BACKUP SLIDES

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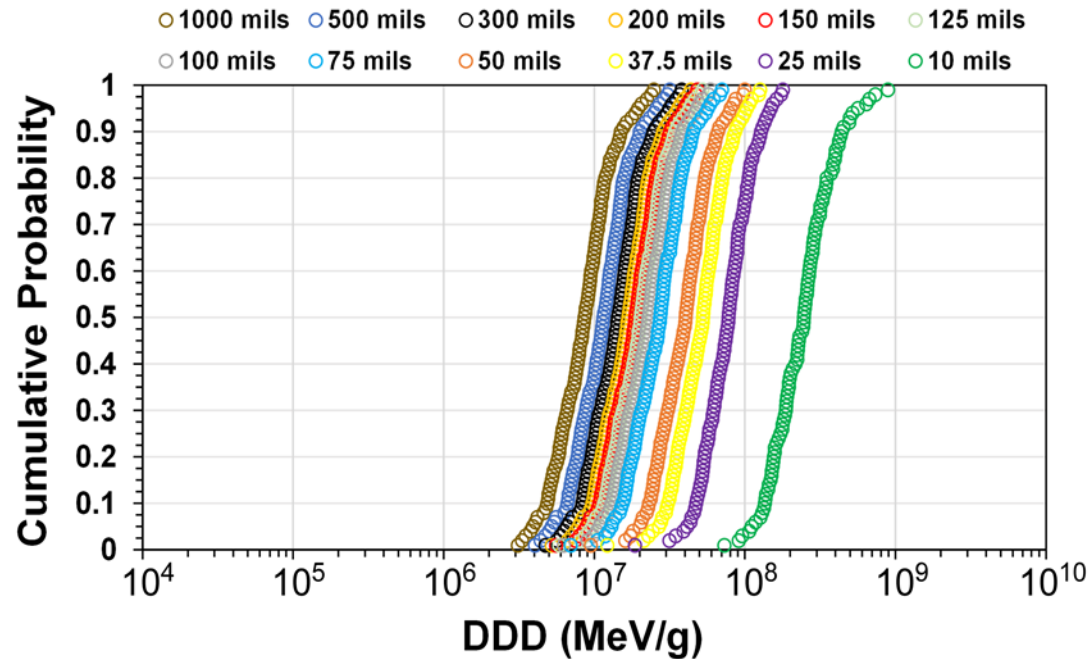
# Device Failure Distribution HV801 Optocoupler



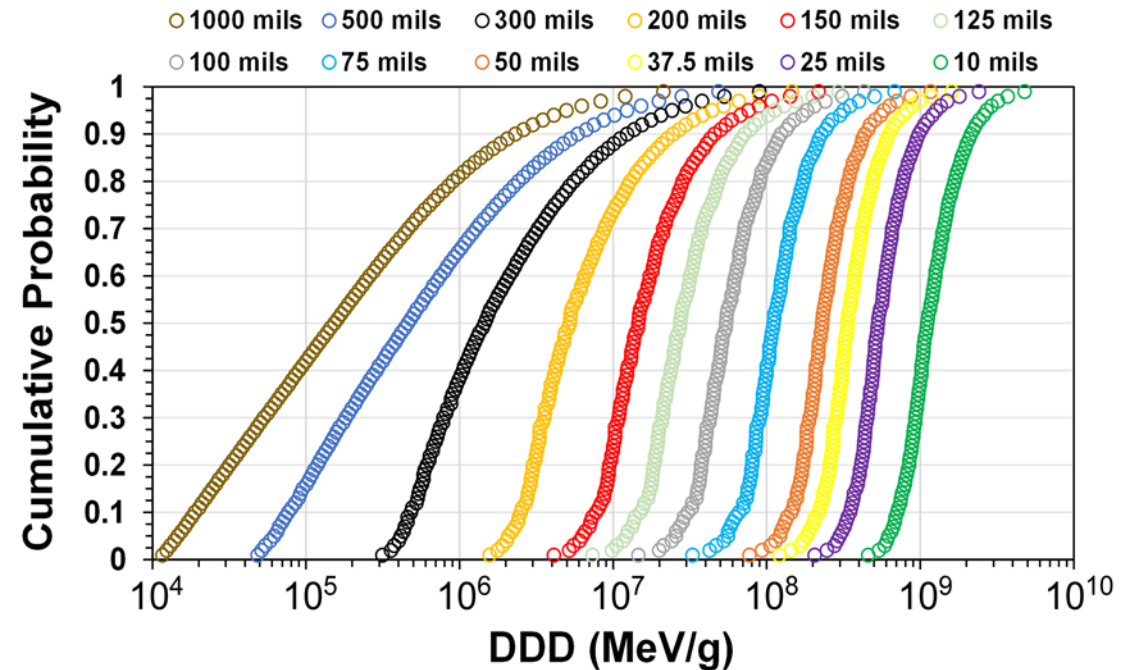
# DDD Probability Distributions for 1 Year 10 – 1000 mils Aluminum



## Low Inclination LEO



## GEO







# Failure Probabilities HV801 Optocoupler

$$P_{fail} = \int [1 - H(x)] \cdot g(x) dx$$

$H(x)$  = CDF for environment dose

$g(x)$  = PDF for device failure

Failure probability ( $P_{fail}$ ) is the probability of a total dose failure during a mission

