Testing with the Laser-Enhanced Arc Jet (IHF) Facility at NASA Ames Research Center

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Presented at the 2018 National Space and Missiles Materials Symposium

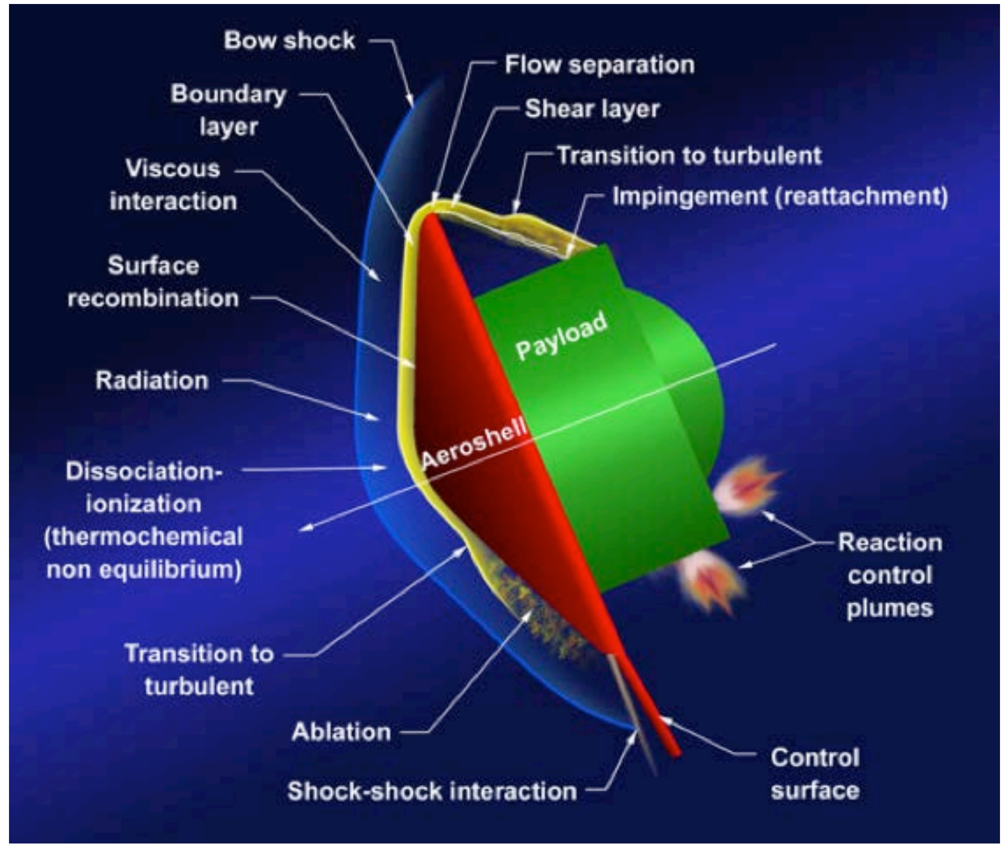
Madison, WI June 25, 2018

Outline

- Background
- Requirements
- System Details
- Key Results
  - Calibration & Ablative TPS Tests
- Concluding Remarks

Background

- During entry, shock-layer radiation and the impact on TPS is important for a number of NASA missions
  - Apollo (Lunar Return)
  - Galileo Probe into Jupiter
  - P-V
  - Stardust
  - Future in-situ robotic missions
    - Venus
    - Sample Return Missions
      - (Mars, Comets and Asteroids)
    - Mars Entry
    - Titan Missions



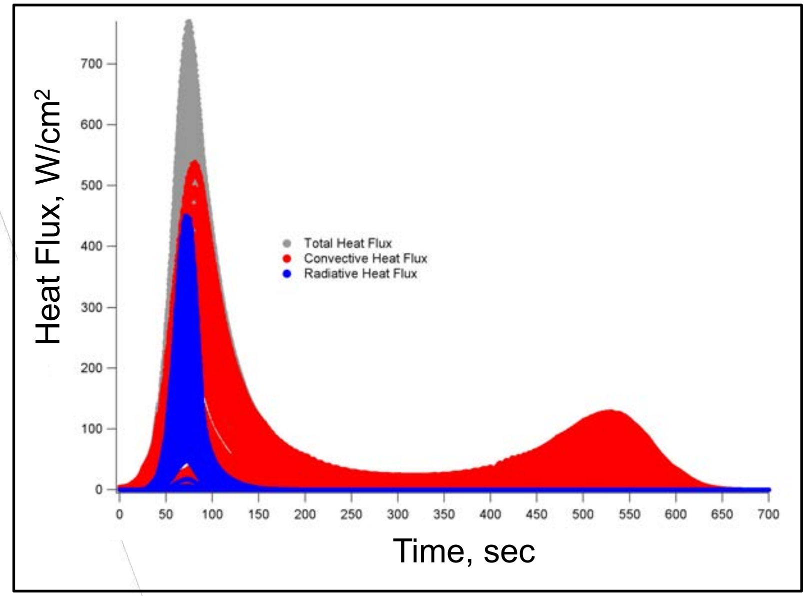
- Near term driver
  - Orion – Lunar return

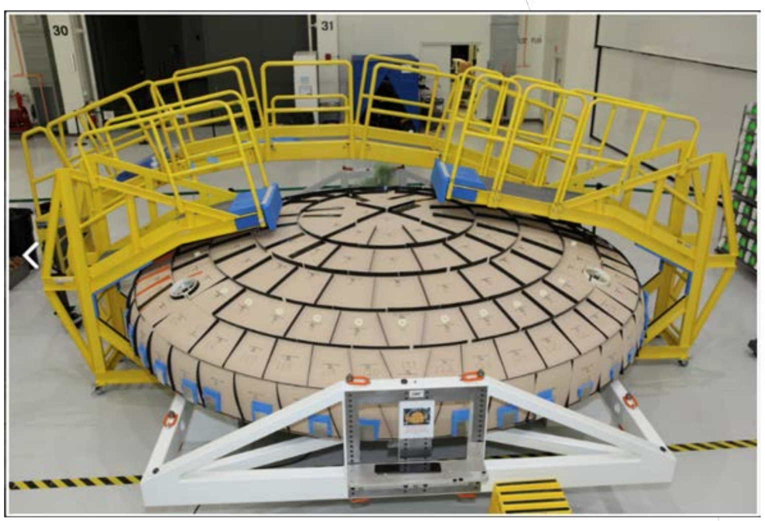
- **Shock layer radiation is a significant percentage of entry heating**
  - **Understanding the ablative TPS material/system response**
  - **Designing and verifying adequate margin**

Requirements:
Orion TPS Certification and Mission Assurance

- Orion Heat shield design
  - EM1 & EM2
  - EM2 certification
- Heat Shield System Certification Challenges
  - Tiled System with gap-filler
  - Compression-pad region

Representative Entry Environment (Lunar Return)



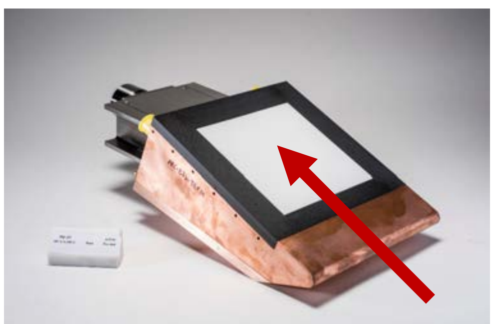


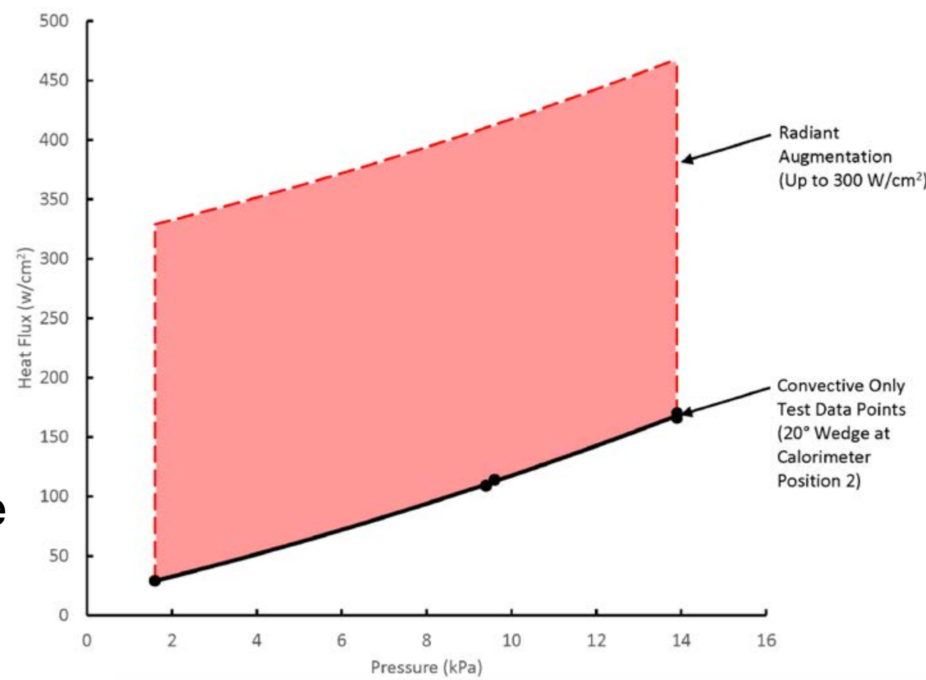
Orion Lunar Capable Heat Shield (Avcoat Tiles)

Crew and Service Module Attachment (Compression Pad with Tension Ties)

Laser Enhancement System Requirements

- Laser enhancement system is designed to add radiant heating to the IHF (Interaction Heating Facility) at NASA Ames Research Center
- Test article configurations
  - Wedge (6”x6”) in a conical nozzle

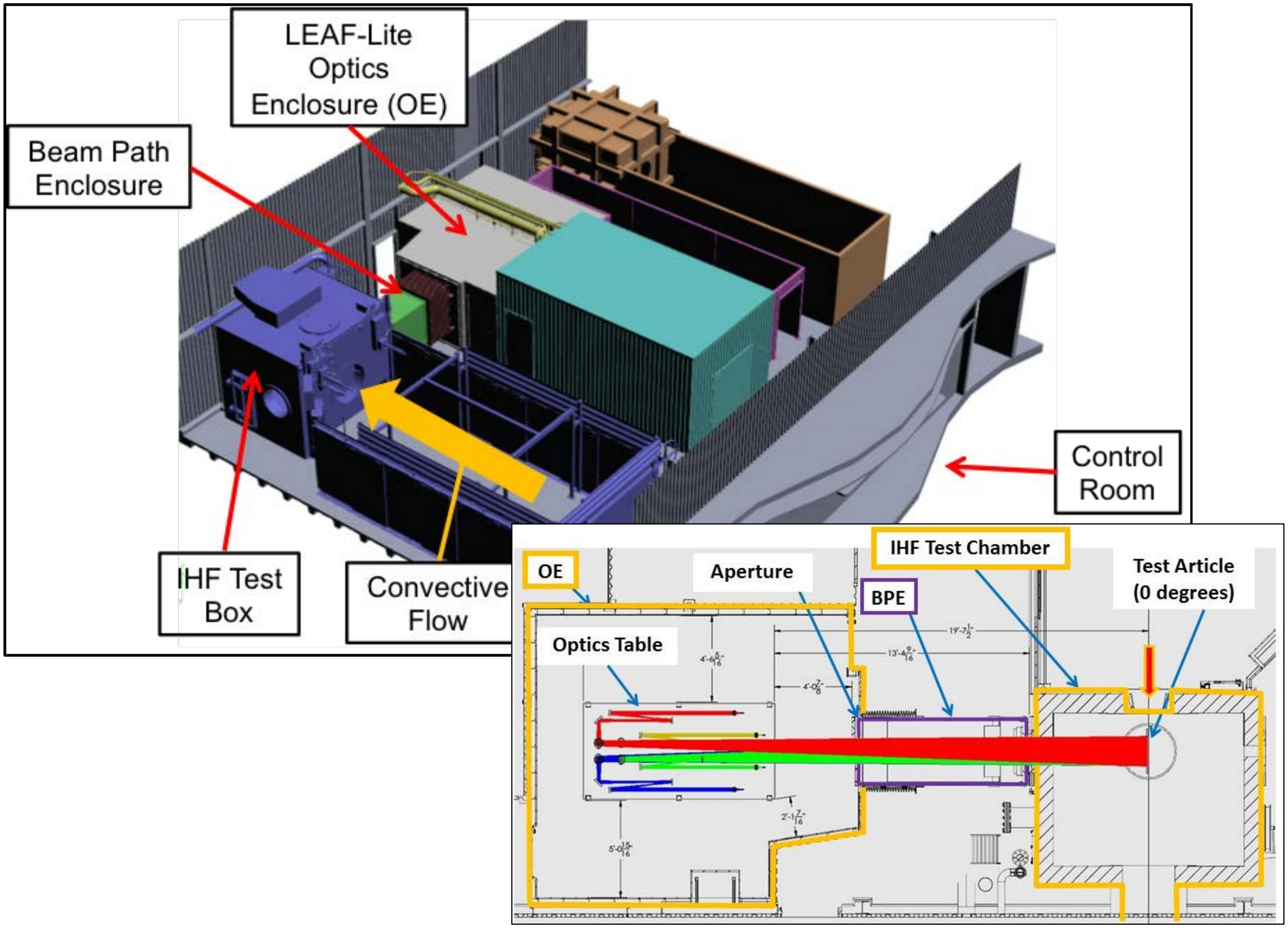




  - Panel (17’x17”)
    - In a semi-elliptic nozzle

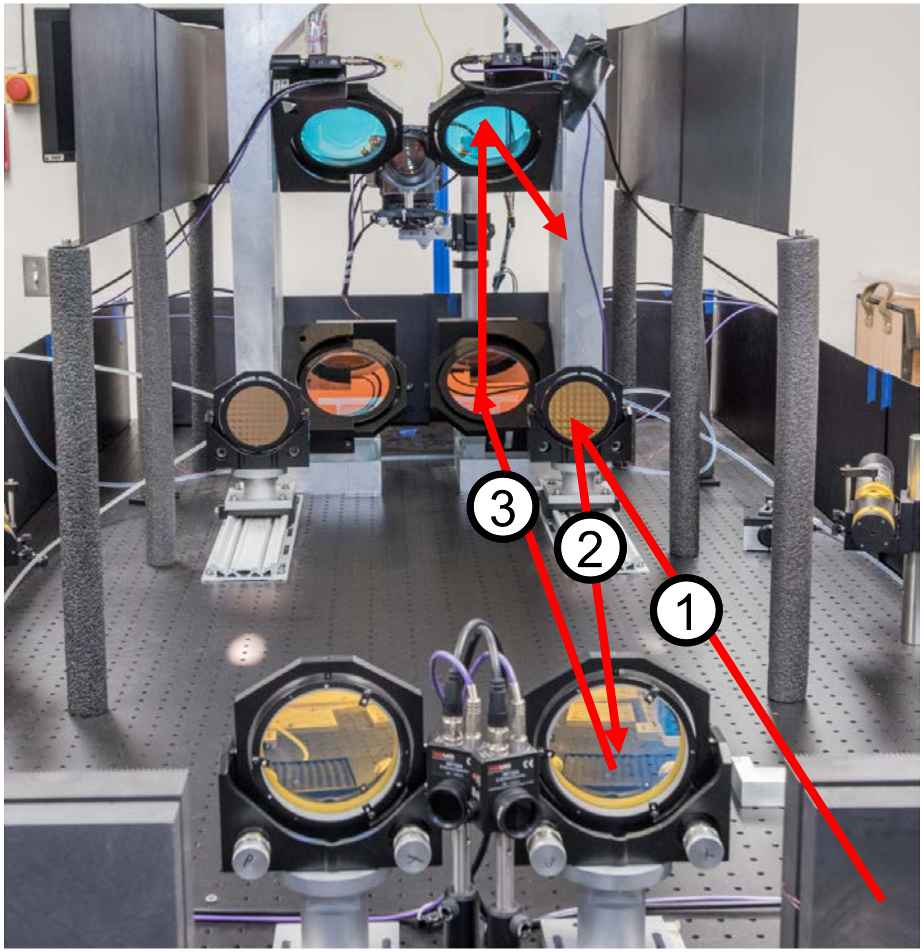
Laser Count	(6” x 6”) Square (W/cm²)	(17” x 17”) Square (W/cm²)
50 kW	195	27
100 kW	390	54
150 kW		80
200 kW		107

Integrated IHF and Laser Enhancement Setup

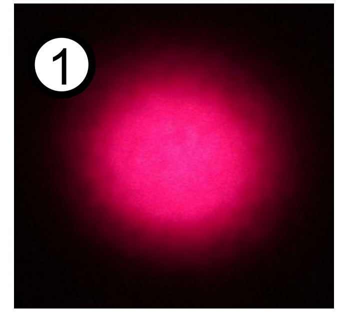


- Major facility upgrades, in addition to the laser power system, include modifications to the plenum, new nozzle (9”), large wedges and overall operational safety.

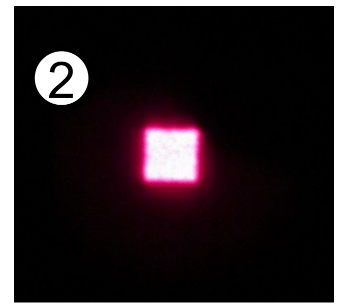
Laser Enhancement Optical Setup



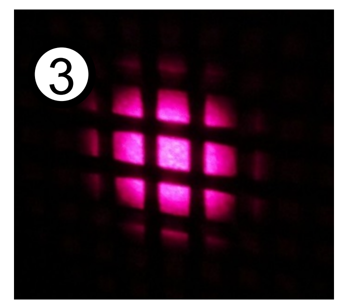
1) Gaussian beam emerges from collimator



2) Beam at the focus of the integrator (1cm x 1cm square spot)



3) Converging beamlets to be reimaged



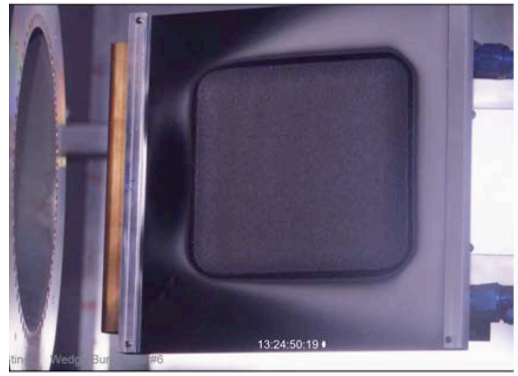
*Images of red guide beam

Laser Enhancement System
Explained by the Lead Engineer

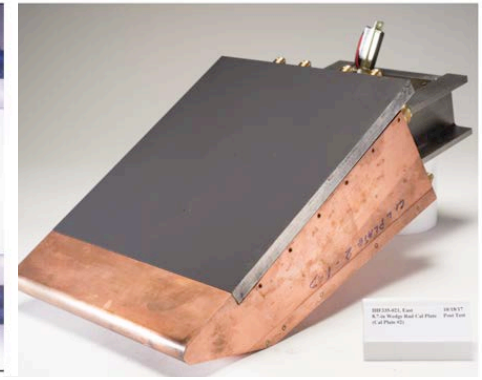
Integrated System Verification and Avcoat Tests

- **Purpose:**
  - Test wedge configuration (6” x 6”)
  - Verify low variation in irradiance
  - Evaluate max heat flux
  - Nearly 40 Tests
- **Tested:**
  - Burn Plates
  - Cal Plates
    - Conv. Cal Plate 6 Gardon Gauges and 3 Pressure Ports
    - Rad. Cal plate - only Gardon Gauges
  - Heatshield materials
    - RCG Coated Tiles (non-ablative)
    - Avcoat – Ablative - Orion
- **Successful with no major problems.**

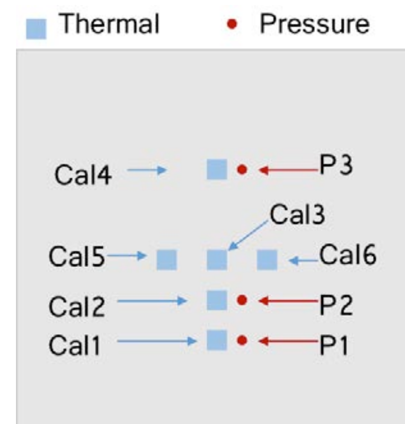
R3 Burn Plate

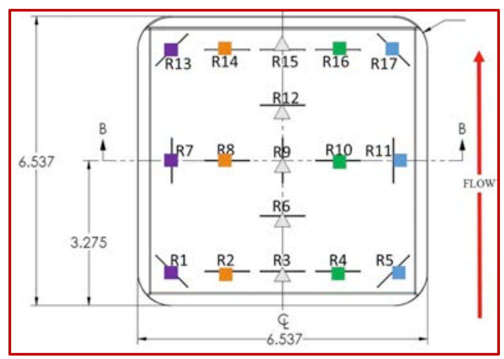


Radiative Cal Plate

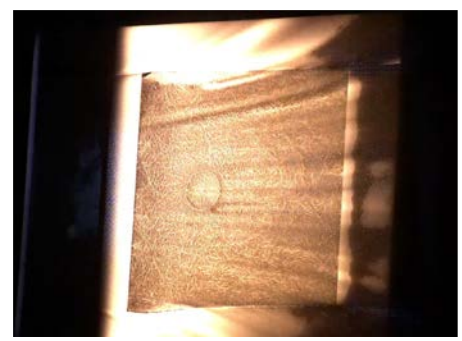


Convective Cal Plate



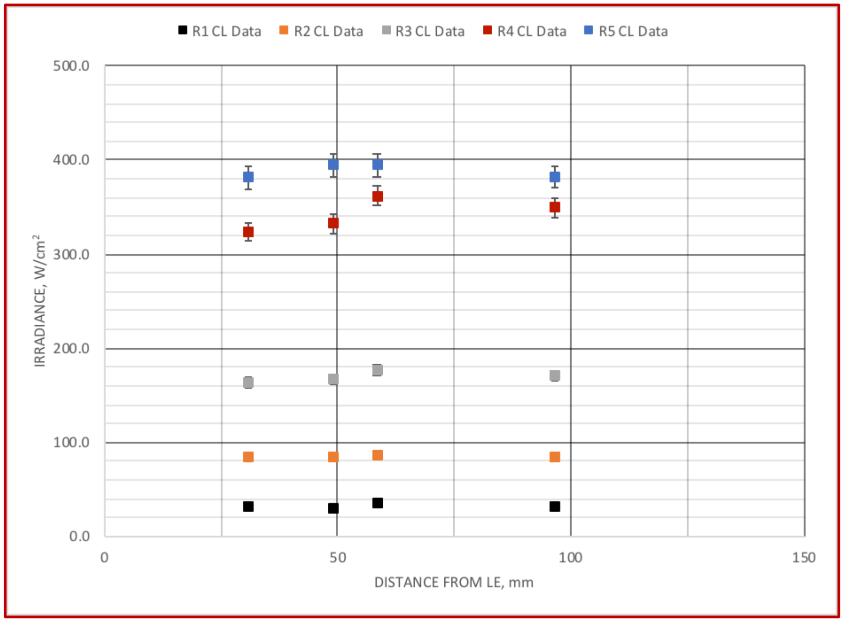


RCG L1-2200 coated plate with 17 near-surface TCs.

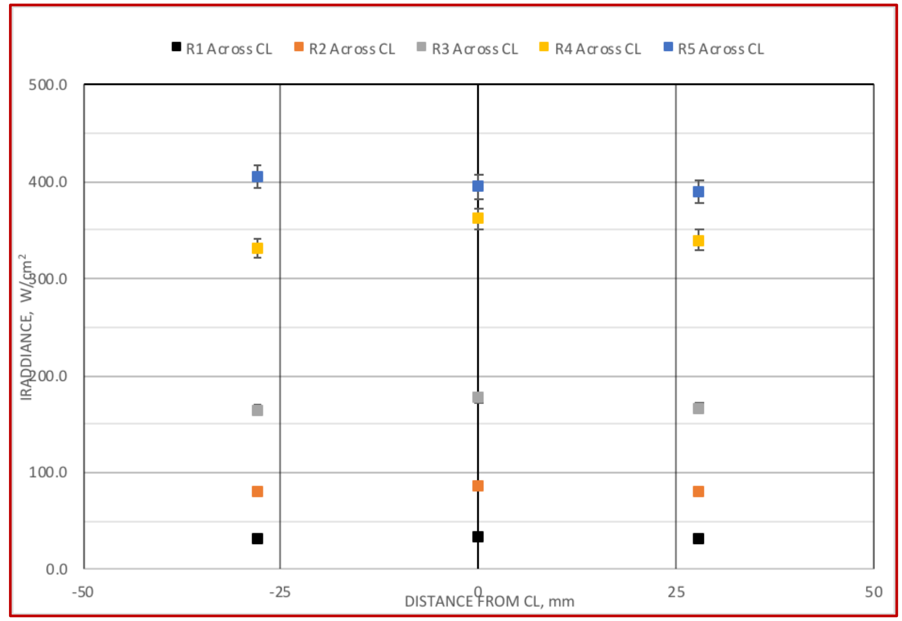


Avcoat

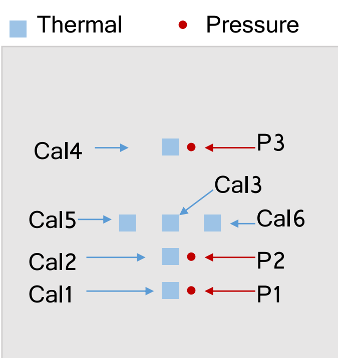
Radiative Heating Calibration Results



Radiative calibration along centerline



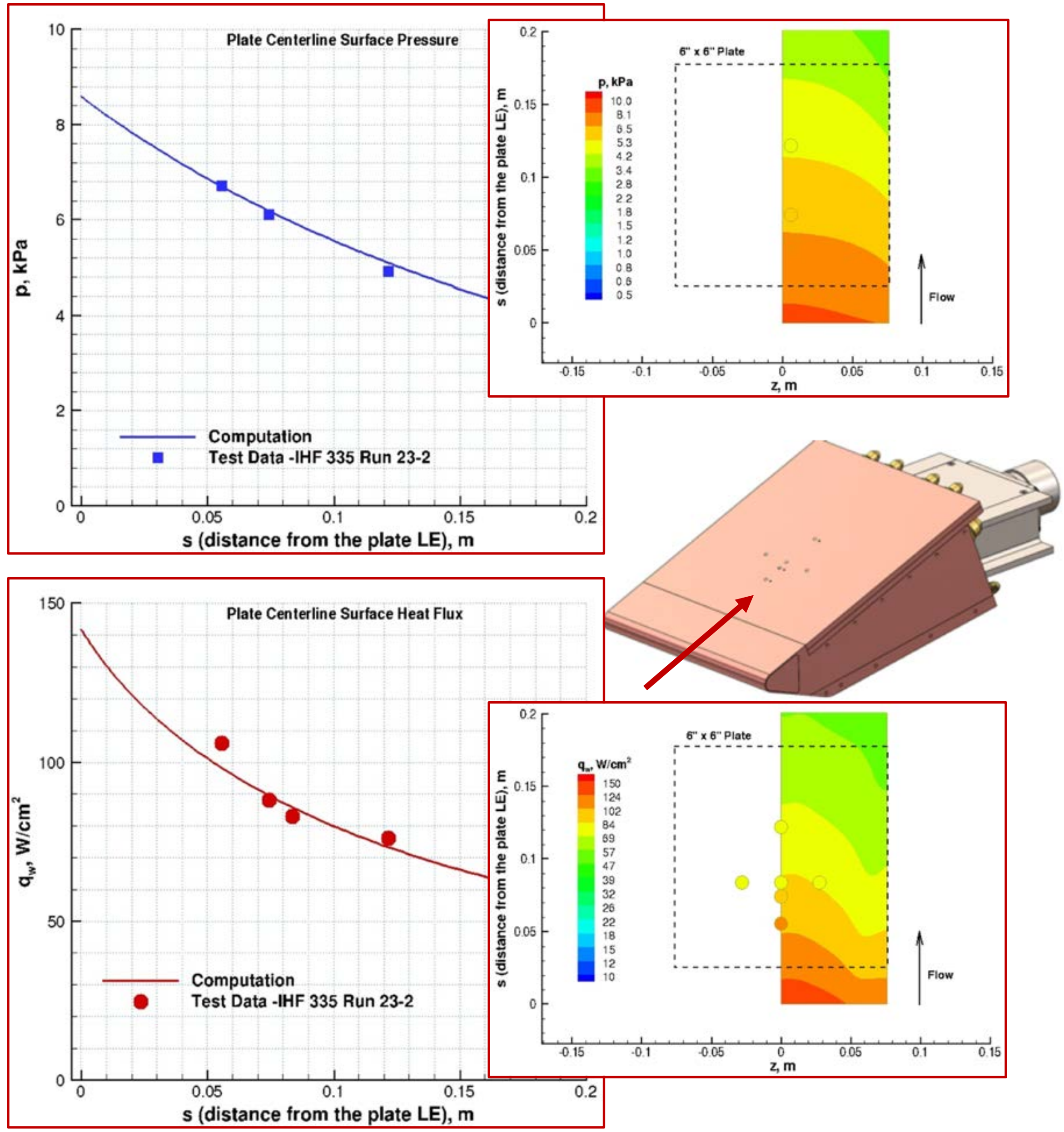
Radiative calibration across centerline



- **Beam is uniform within 6% of the average irradiance across all conditions.**
- **Measurements across multiple runs, for R3 and R4, run-to-run variability is <11%.**

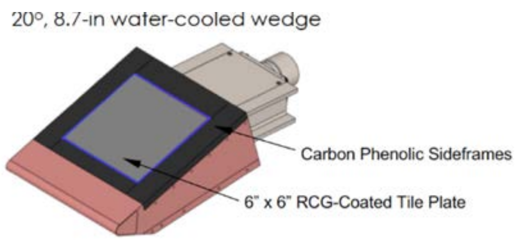
Convective Heat Flux Calibration Results & Comparisons with CFD

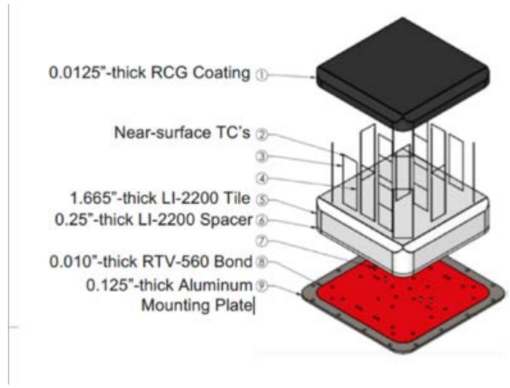
- Convective pressure and heat-flux measured were compared with CFD
- As predicted, the heat-flux and pressure decrease with increasing distance from leading edge
- The comparison shows the measurement and CFD are in agreement

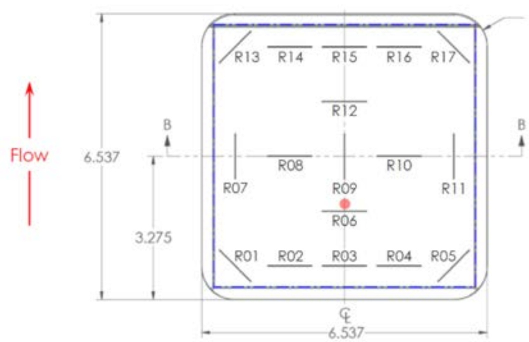


CFD Simulation in support of Orion : Dr. Gockcen

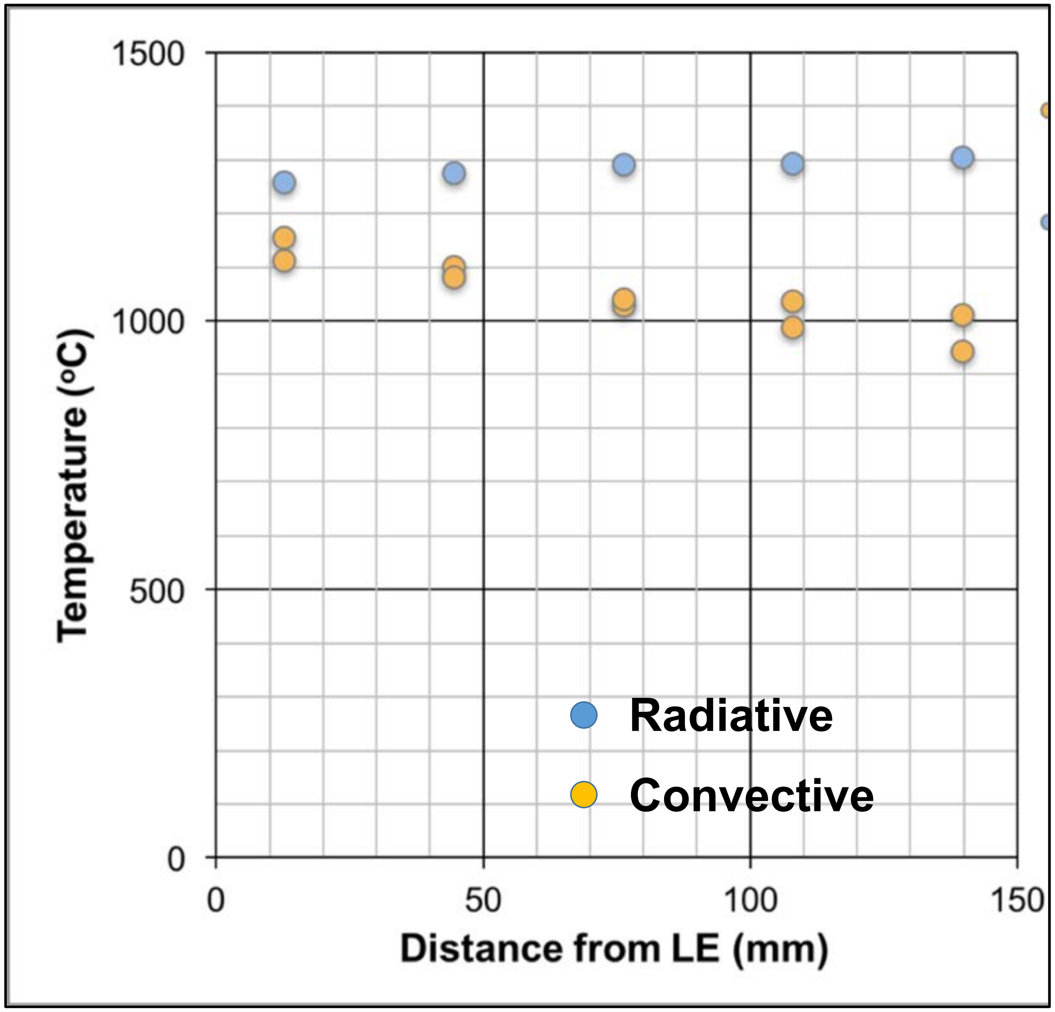
RCG Coated Tile Test Results





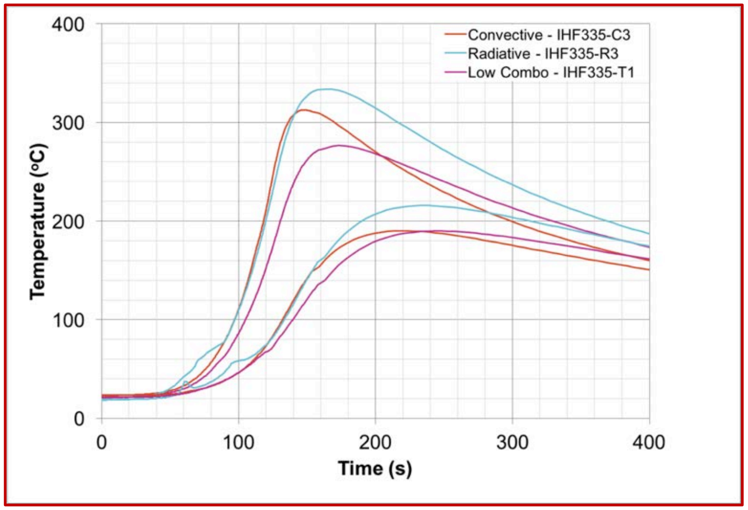


Centerline Data

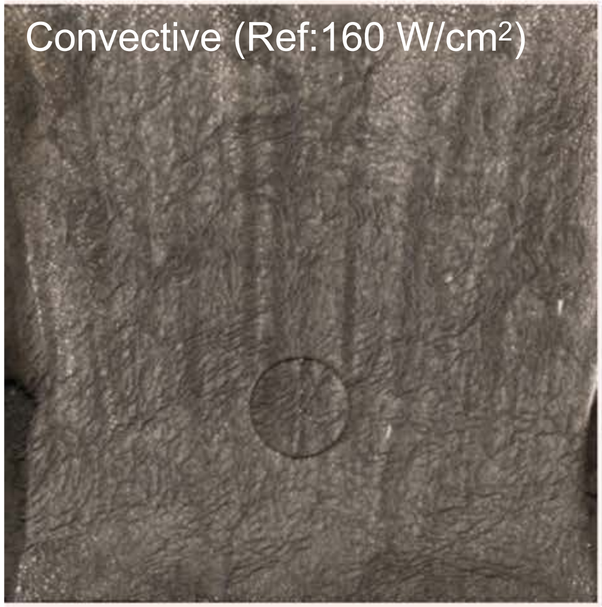


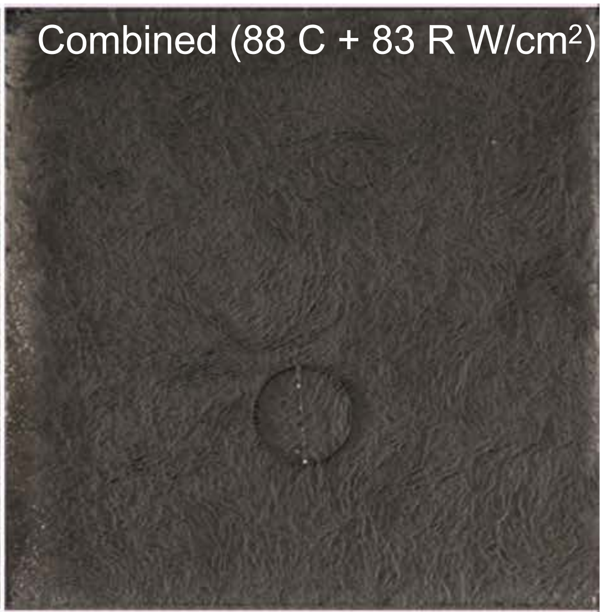
- **Near Surface Thermocouple on RCG coated test article captures the trend observed with convective and radiative cal plates**
  - Run at lowest convective (and radiative conditions

Avcoat Test Results



- Avcoat test results show differences between radiative, convective and combined heating





• Entire surface covered in glass	• Glass limited to periphery	• Glass limited to periphery

Near- and Longer-Term Use

- Near-term focus is to support Orion
  - EM2 certification using the combined convective and radiative heating capability

Near Term Radiative Capability

Laser Count	(6" x 6") Square (W/cm^2)	(17" x 17") Square (W/cm^2)
50 kW	**195**	**27**
100 kW	**390**	**54**
150 kW		**80**
200 kW		**107**

- Longer-term use by both NASA and other customers envisioned.
  - The shock layer radiation for most of planetary entry missions, with the exception of Jupiter, are below 1000 W/cm2
  - Testing at higher heat-flux on a reasonably size articles could be achieved (with some facility and optical system modifications).
    - 200 kW system on a 6" x 6" article (> 700 W/cm2 radiative)
    - Testing in vacuum with radiative heating alone can provide insight into material behavior

Concluding Remarks

- Laser Enhanced IHF - a unique capability is nearing completion
  - Combined radiative and convective testing capability will be used primarily for Orion in the near term.
  - 100 kW system has been successfully installed, operational
  - System upgrade in progress will bring the 200 kW capability to the IHF
- Testing
  - Calibration of the laser enhanced IHF is completed
  - Exploratory testing of Avcoat is completed
  - Orion TPS testing in support of EM1 and EM2 will begin soon.

Acknowledgement & Contact Information

- Thanks to the staff of the NASA Ames Entry Systems and Technology Division that has contributed to the development of this new capability

- Thanks to the Orion Program Office for funding this expanded testing capability for future crewed, and un-crewed, missions

POC:

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Avcoat: Arc jet flow on – Laser Off
Convective Heating Only

Avcoat: Arc jet flow off – Laser On Radiative Heating Only

Avcoat: Arc jet flow on – Laser on Combined Heating