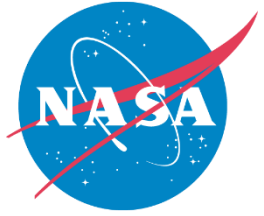


Initial testing of 10" ducted propeller in hover

Nicole Pettingill, Noah Schiller, Nikolas Zawodny, Matthew Galles
NASA Langley Research Center
In collaboration with ONERA

Spring 2021 Acoustics Technical Working Group VIRTUAL Meeting
April 14, 2021

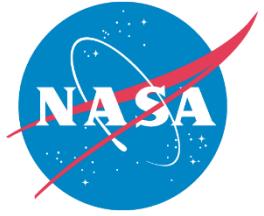
Outline



- Introduction
- Duct design and printing
- Facility and hardware set up
- Preliminary performance and acoustic data
- Future Work

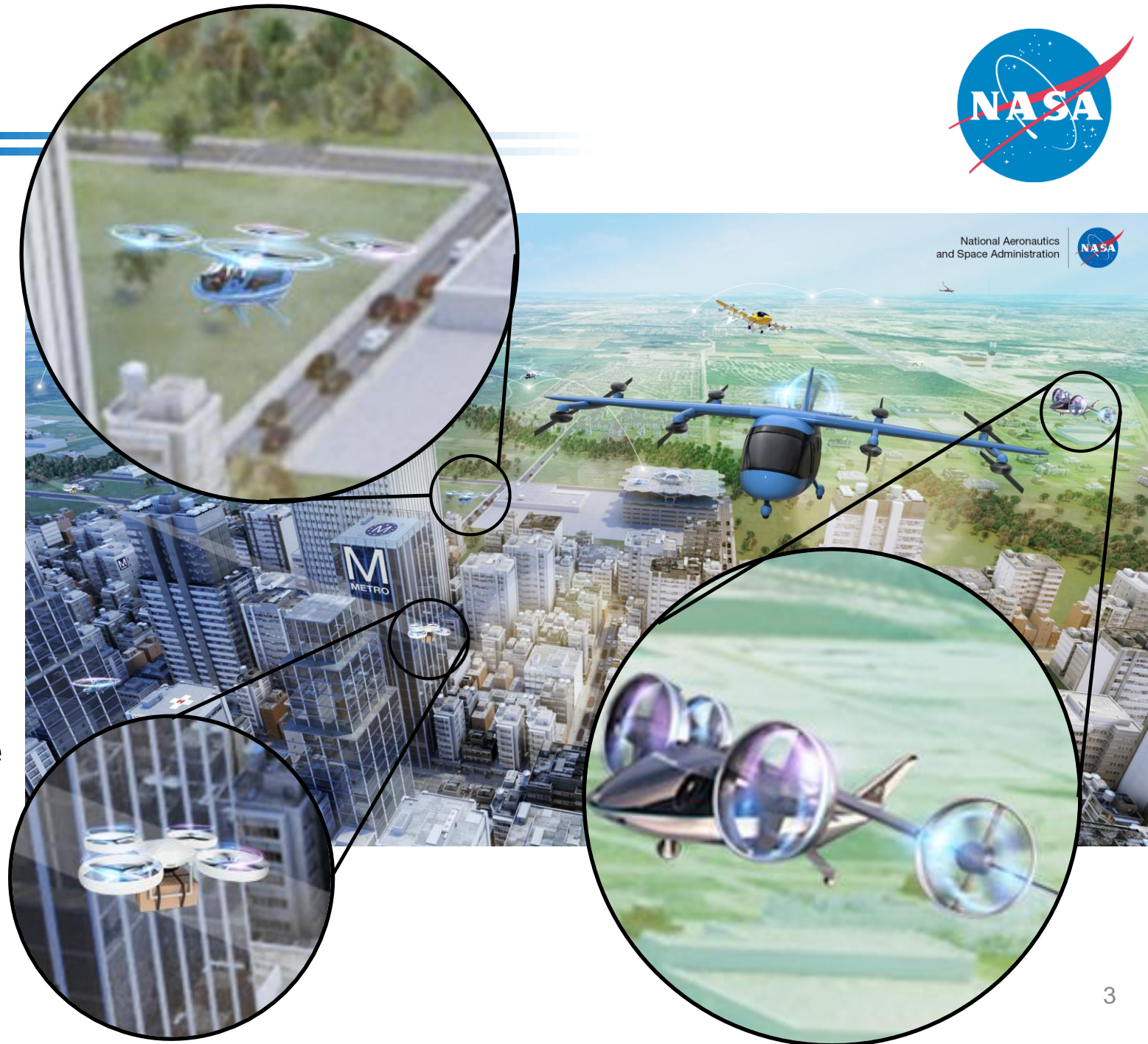


Introduction



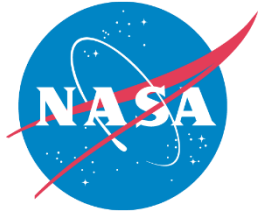
Objective: to help assess benefits of acoustic liners installed in a scaled duct

- Many AAM aircraft configurations are being considered, some of which have ducted propulsors
- Previous work on liner design and prediction have been done*
- This presentation will focus on the test setup and initial results



* Simon, F. and Schiller, N., "ONERA/NASA Collaboration on Noise Reduction for Ducted Rotors," NASA Acoustics Technical Working Group Meeting, Apr 8, 2020.

Design and set up



- Blade(s)
 - 9.6" diameter, 3-bladed rotor at 9,000 RPM
 - Target design thrust = 1.9 lbs
 - Tip pitch angle $\theta = 10^\circ$
 - NACA 0012 airfoil, no twist
 - Blades manufactured via Stereolithography (SLA)

- Ducts
 - Two ducts
 - Untreated hardwall duct
 - Low resistance LEONAR lined duct (L02)
 - Straight ducts, 10" inner diameter, 1.2" thick, 0.6" inlet and exhaust lip radius, 2.4" axial extent (of the straight duct section), blade tip clearance 4% of duct inner radius
 - Ducts manufactured via stereolithography (SLA)

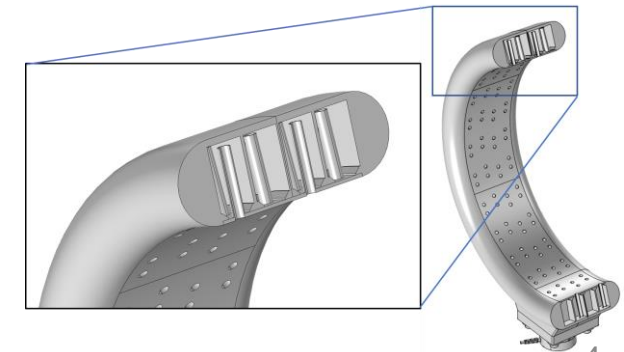
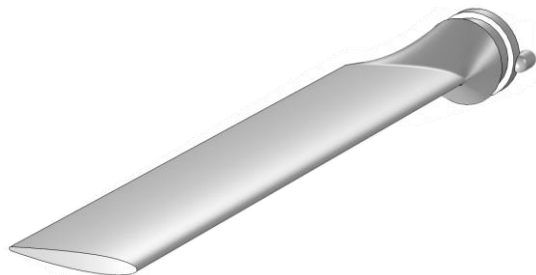
L02 Duct



Hardwall Duct

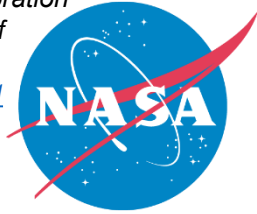


Propeller Blade



Experiment: Facility and Setup

* Whiteside, S. K. S., Zawodny, N. S., Fei, X., Pettingill, N. A., Patterson, M. D., Rothhaar, P. M., "An Exploration of the Performance and Acoustic Characteristics of UAV-Scale Stacked Rotor Configurations", AIAA SciTech 2019, <https://doi.org/10.2514/6.2019-1071>

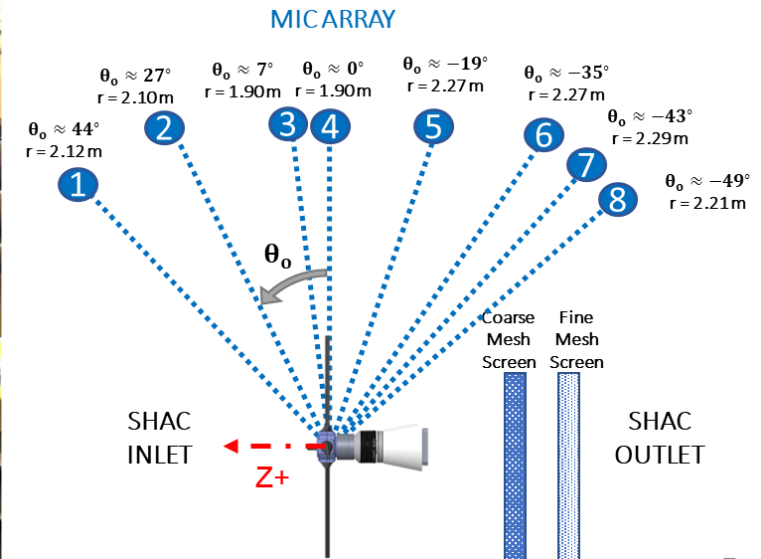
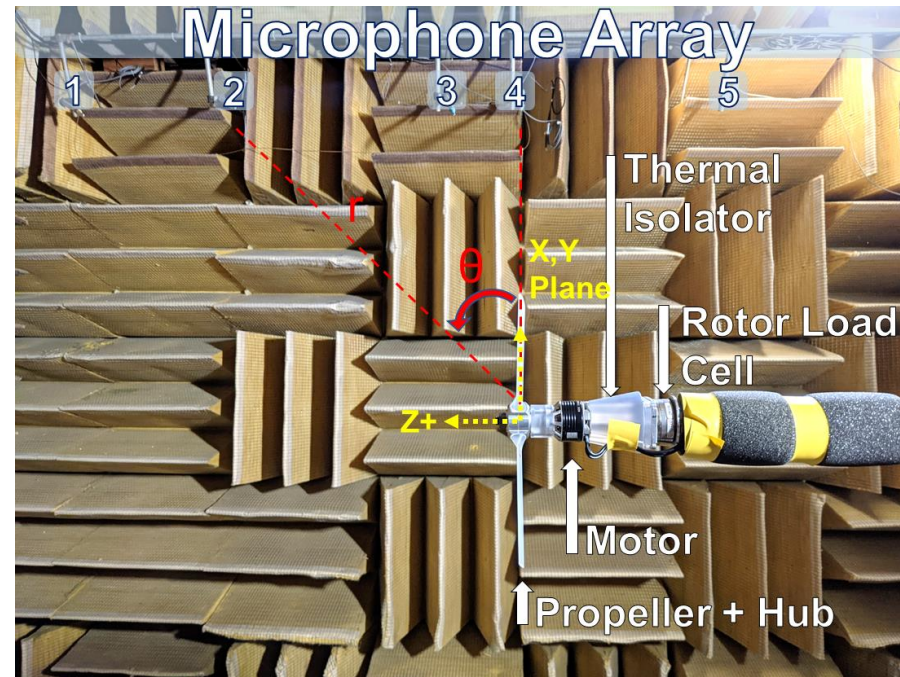


Small Hover Anechoic Chamber (SHAC)*



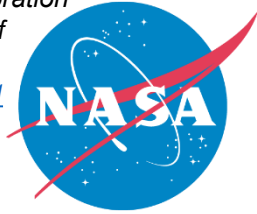
- Room dimensions = [3.87 x 2.56 x 3.26] m
 - Acoustically treated (cutoff down to 250 Hz)
 - Mesh screens reduce the onset of recirculation
- Hardware
 - KDE 2814XF-515 motor
 - Duct mounted on 1" 8020 axial track (~6" below rotor loadcell)

- DAS: Brüel & Kjær (BK) LAN-XI DAQ and BK Connect Software
 - 6 B&K Type 4939 Free-Field microphones + 2 B&K Type 4954B microphones
 - Laser sensor tachometer
 - 2x 6-Component AI-IA mini40 multiaxis load cell
 - Hot Wire Probe + Thermistor

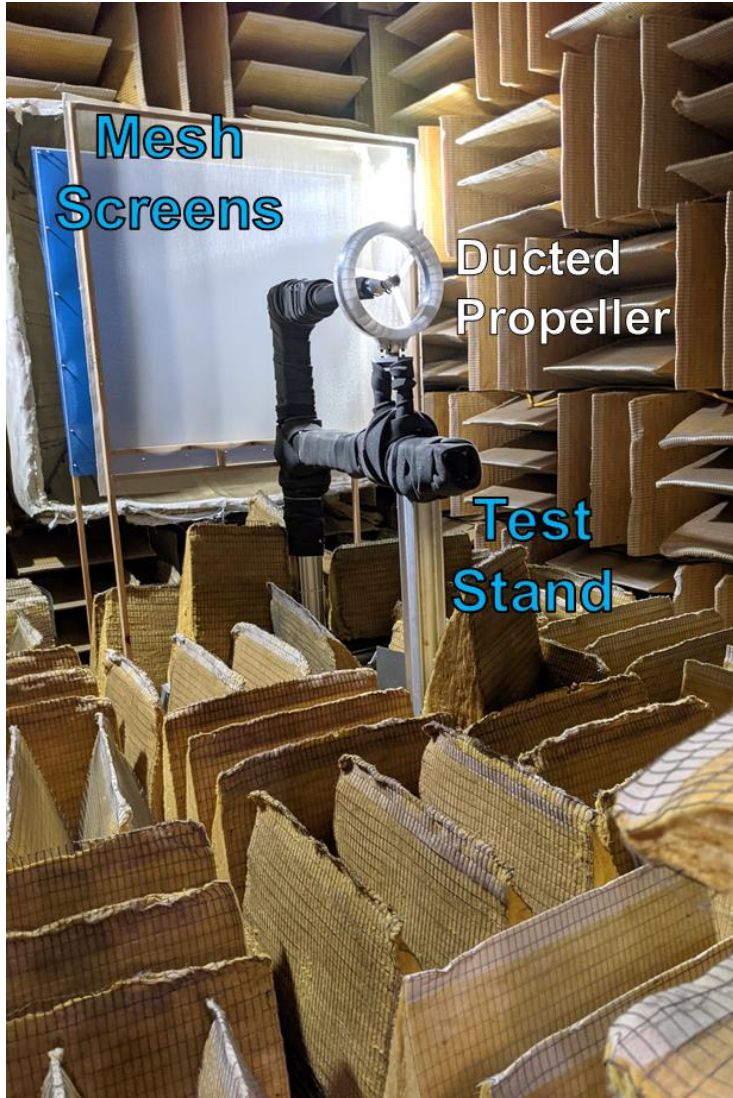


Experiment: Facility and Setup

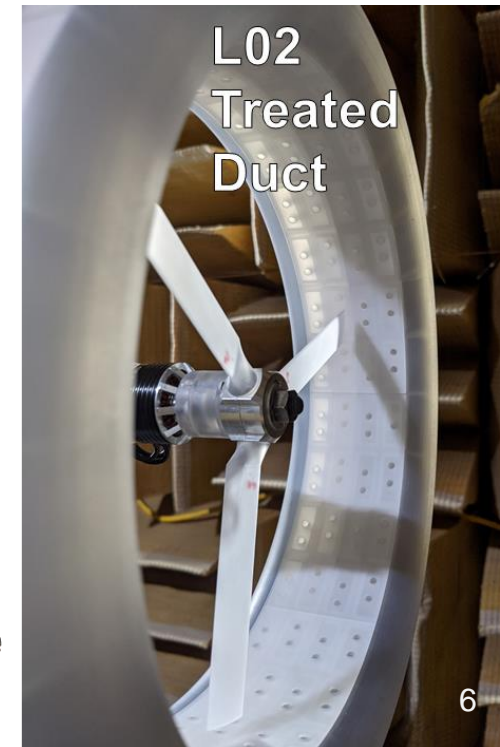
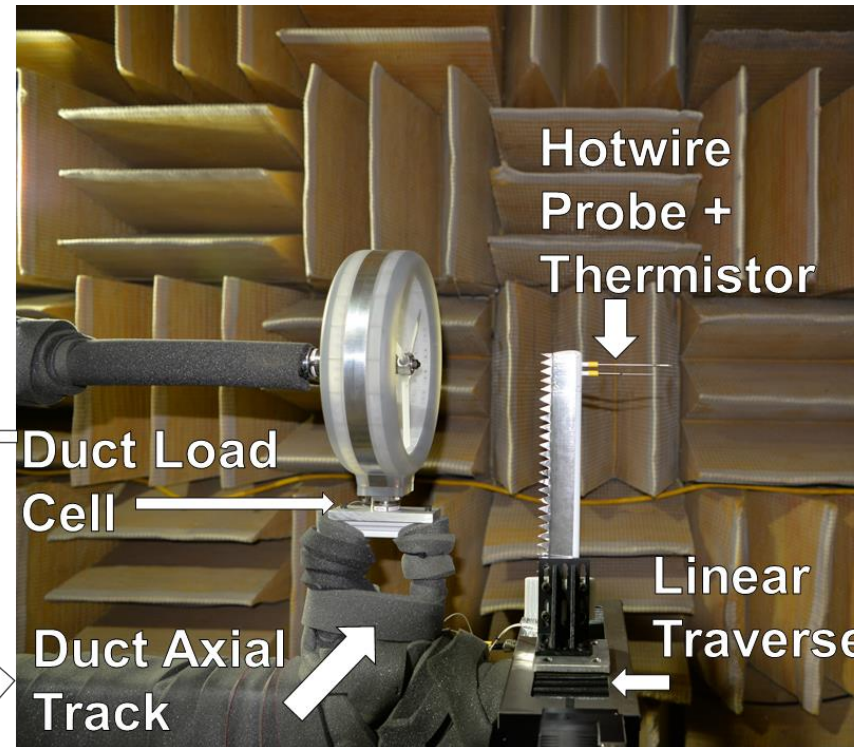
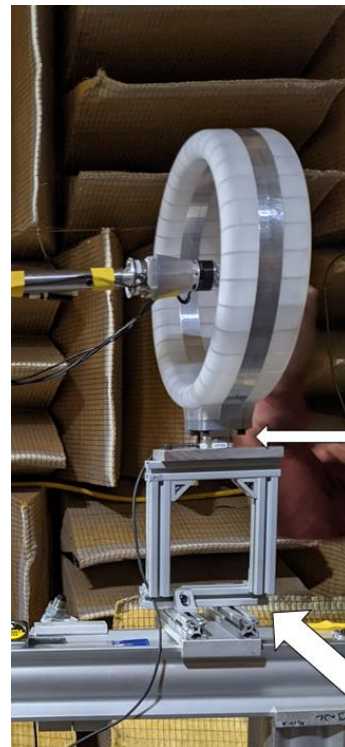
* Whiteside, S. K. S., Zawodny, N. S., Fei, X., Pettingill, N. A., Patterson, M. D., Rothhaar, P. M., "An Exploration of the Performance and Acoustic Characteristics of UAV-Scale Stacked Rotor Configurations", AIAA SciTech 2019, <https://doi.org/10.2514/6.2019-1071>

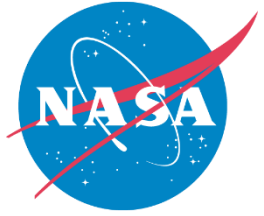


Small Hover Anechoic Chamber (SHAC)*



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 - Laser sensor tachometer
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 - Hot Wire Probe + Thermistor





Testing configurations

Three propeller configurations

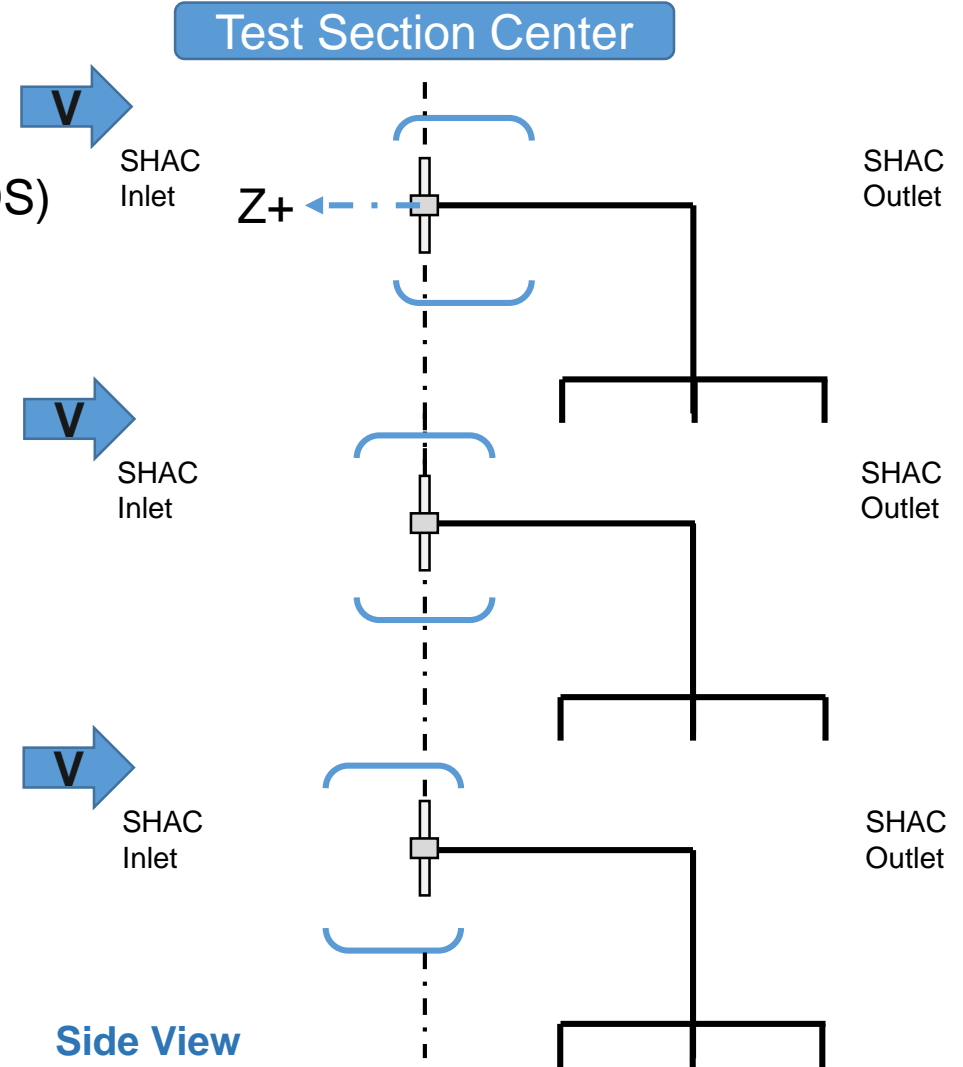
- Isolated propeller (no duct)
- Hardwall untreated ducted propeller
- L02 treated ducted propeller

Two flow conditions

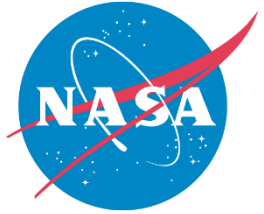
- Background flow on ($V \sim 5$ m/s)
- Background flow off ($V = 0$ m/s)

Three duct positions

- Duct downstream (DS)
- Duct centered
- Duct upstream (US)

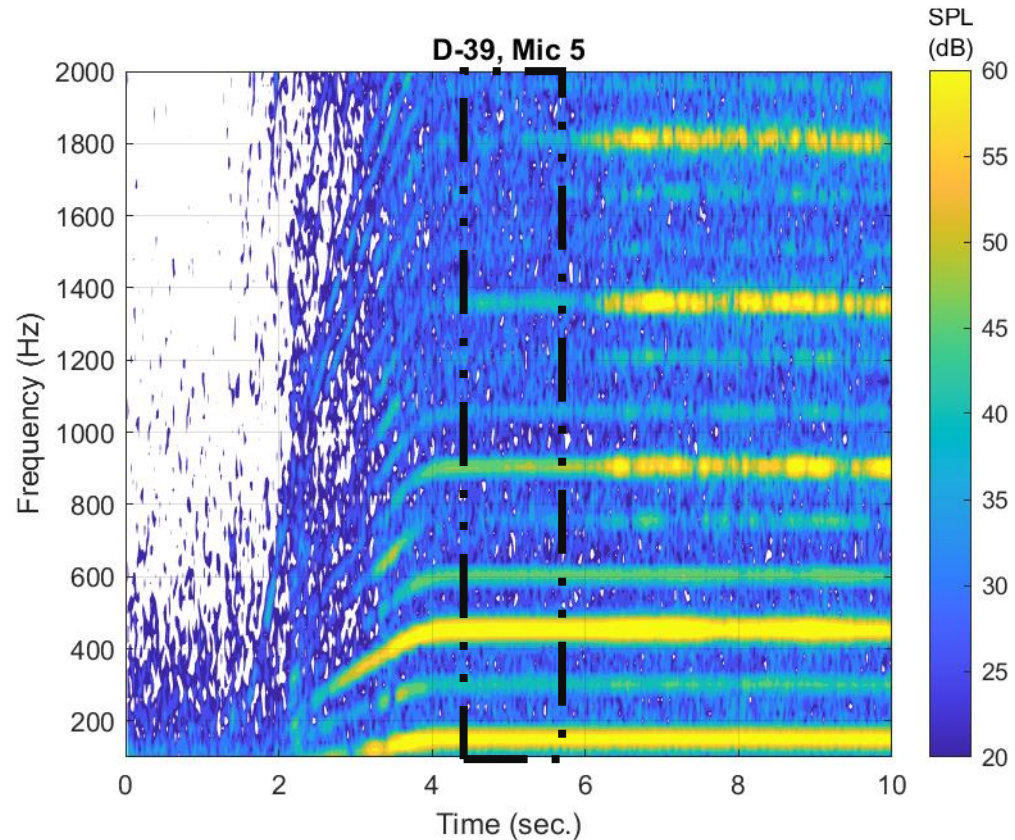


Recirculation in Static Tests – Isolated Propeller



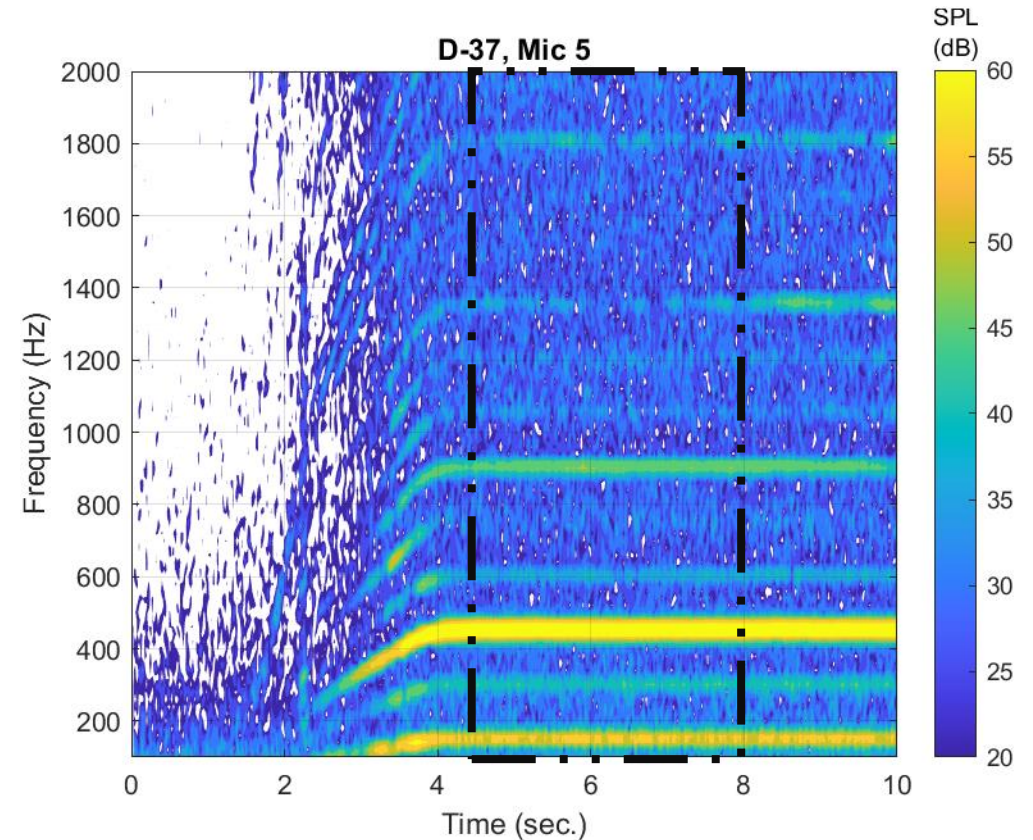
Without Meshscreen Treatment

~1.5 seconds before onset of recirculation

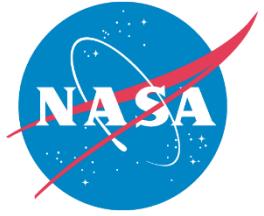


With Meshscreen Treatment

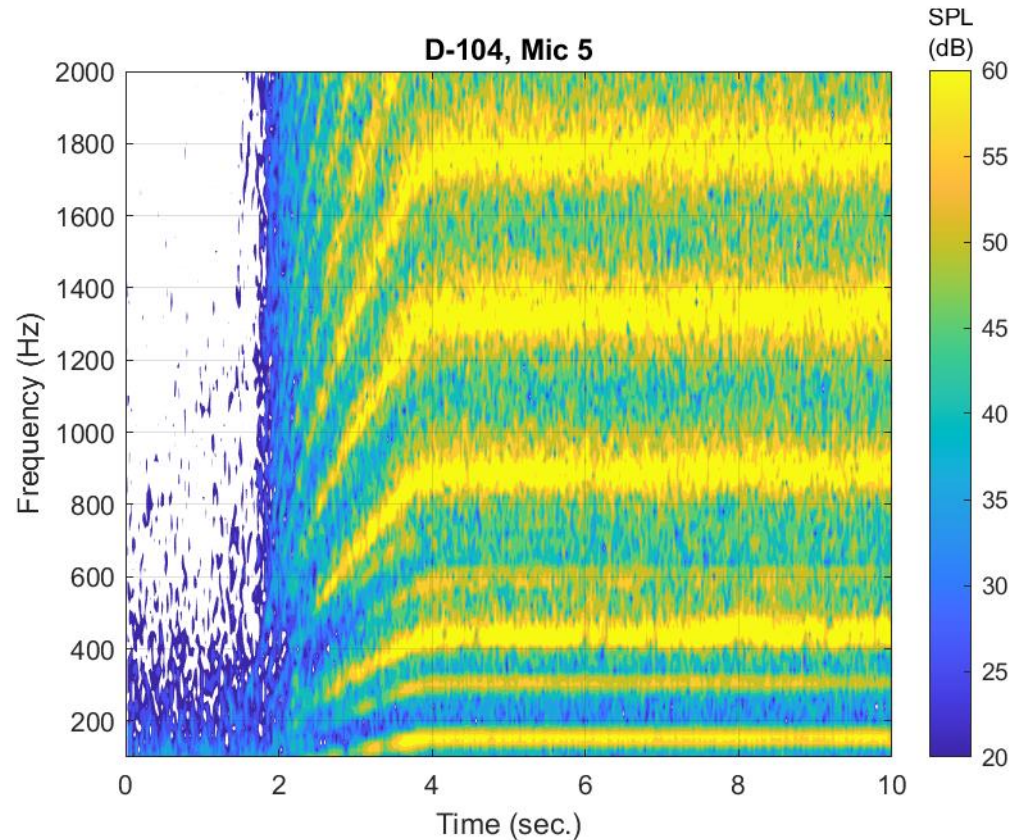
~3.5 seconds before onset of recirculation



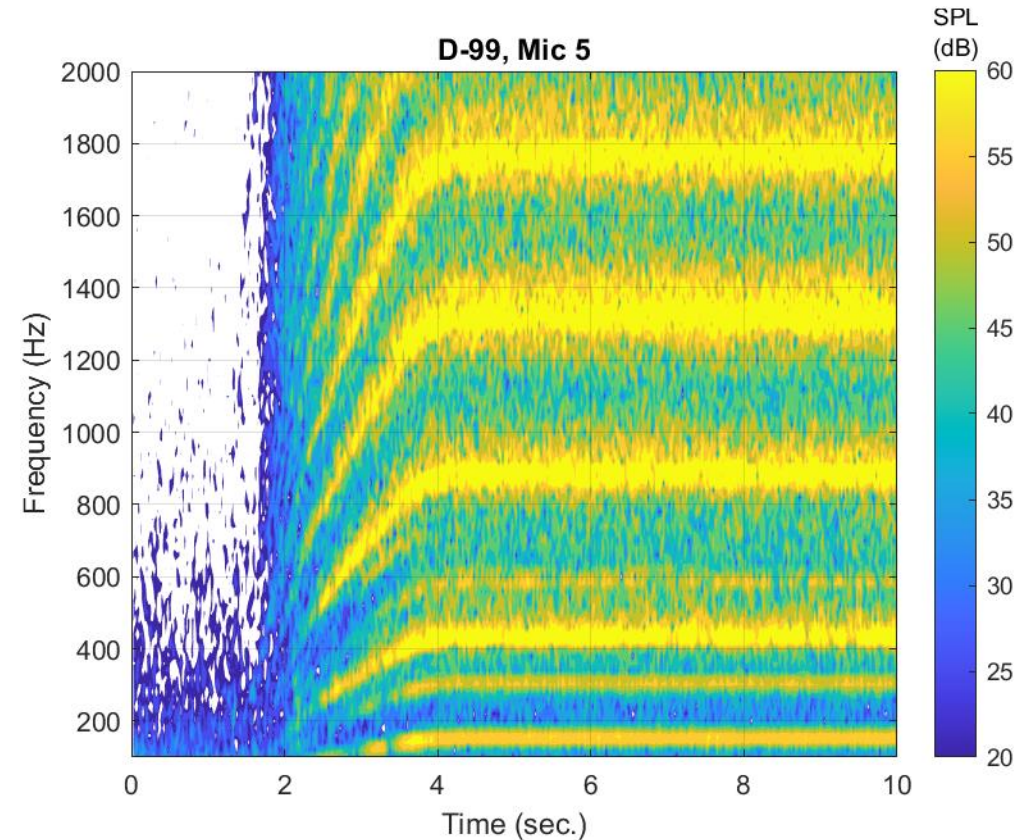
Recirculation in Static Tests – Hardwall Duct



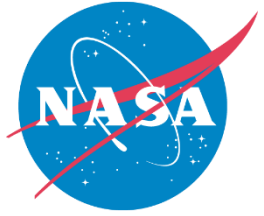
Without Meshscreen Treatment
Onset of recirculation not apparent



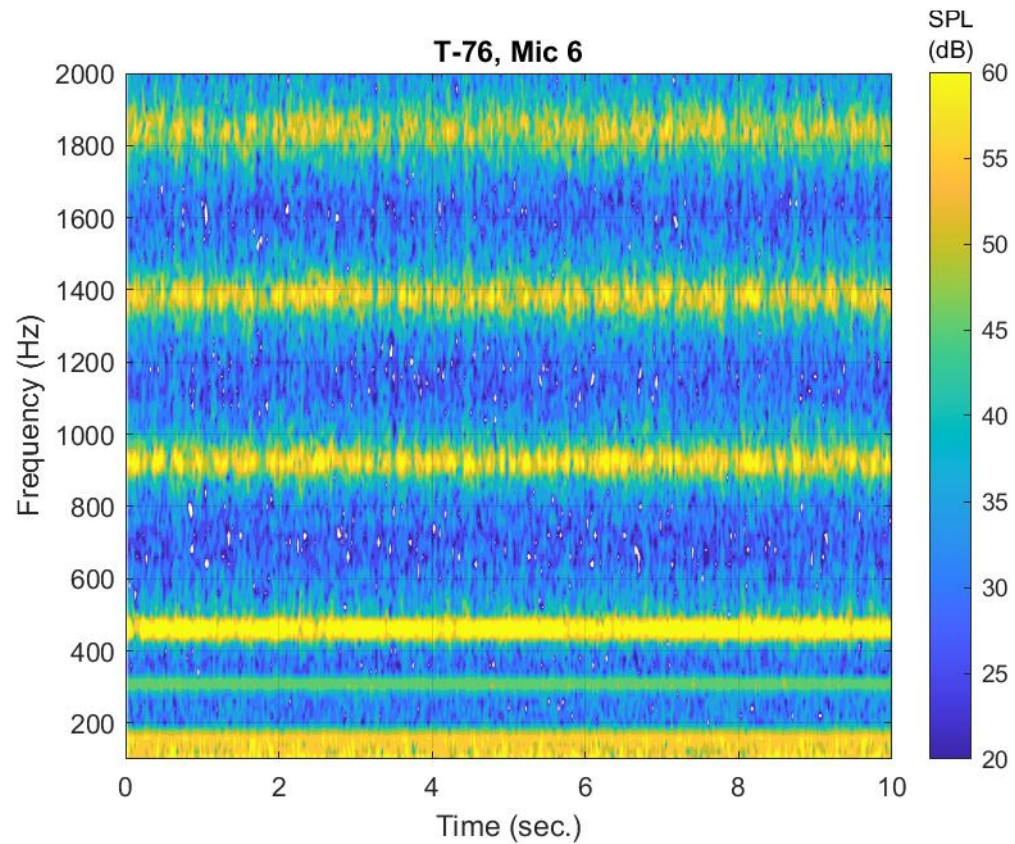
With Meshscreen Treatment
Onset of recirculation not apparent



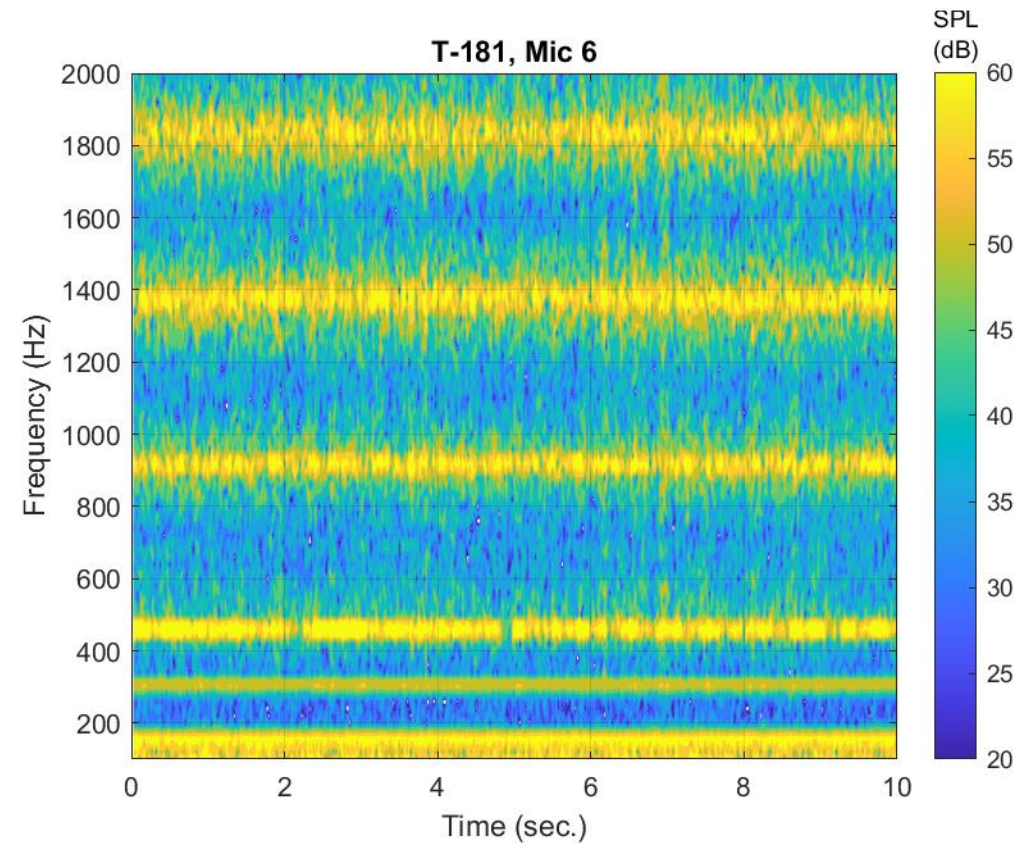
Flow on cases – no recirculation concerns



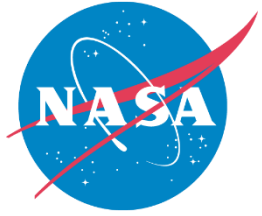
Isolated Propeller



Hardwall Duct

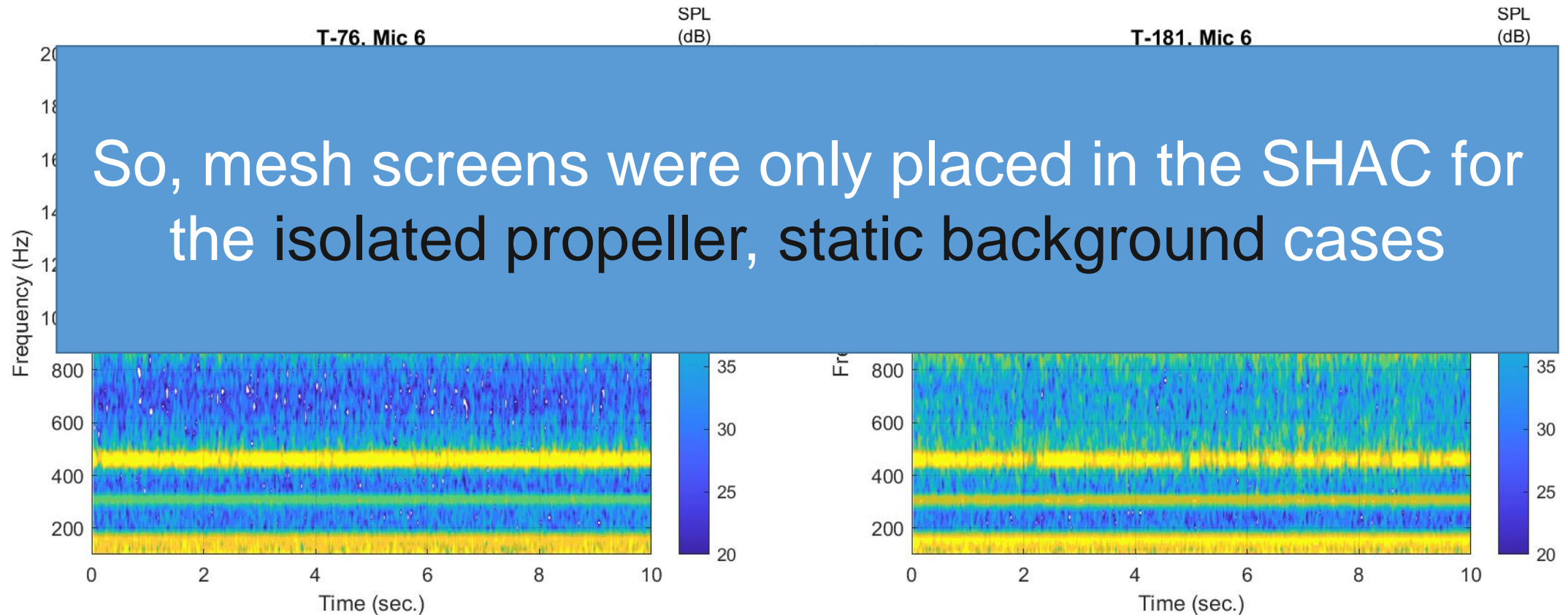


Flow on cases – no recirculation concerns

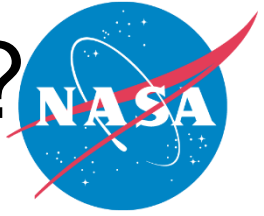


Isolated Propeller

Hardwall Duct

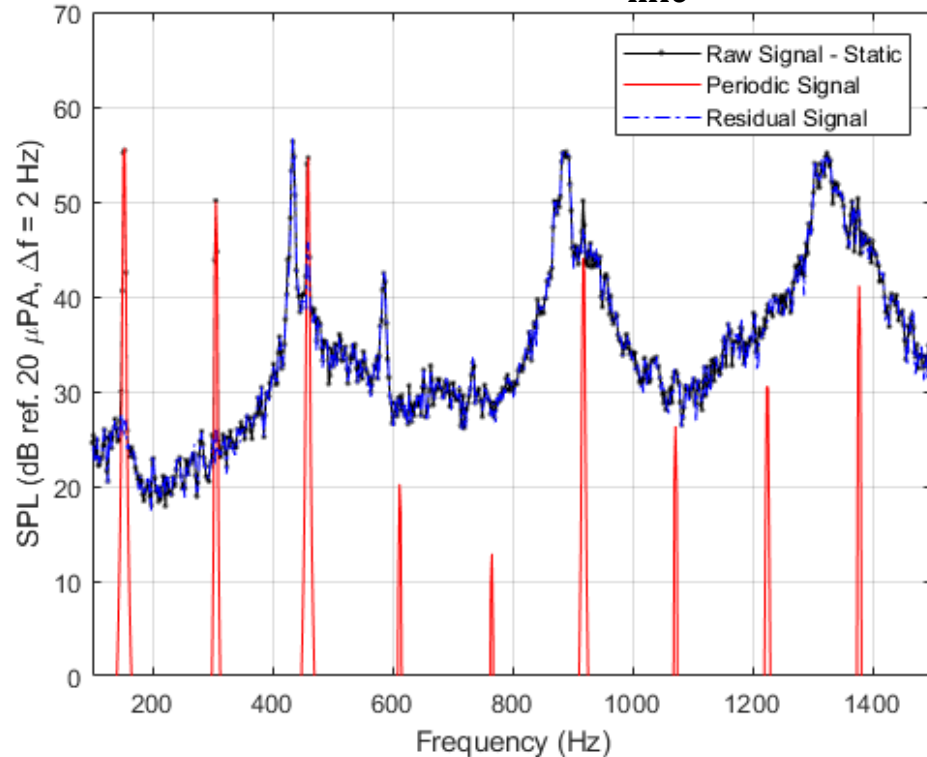


Why have the background free stream flow on?



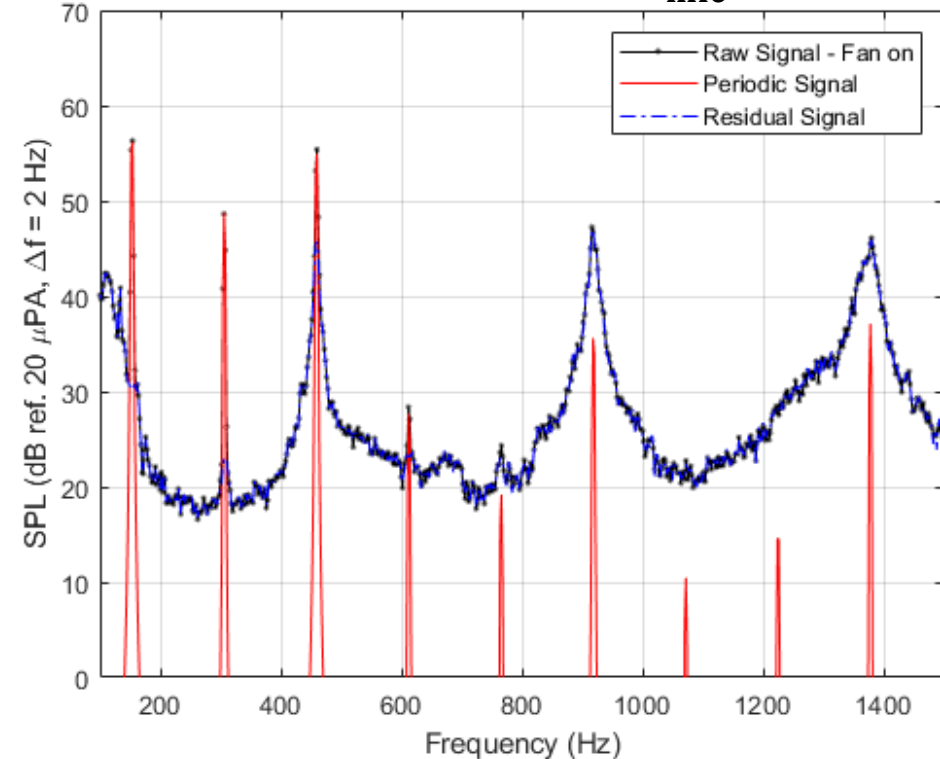
Hardwall Duct US

Static, 9198 RPM, $\theta_{mic} = 0^\circ$



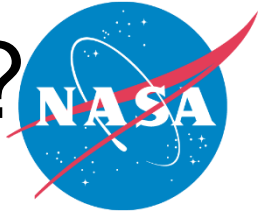
Hardwall Duct US

Flow On, 9198 RPM, $\theta_{mic} = 0^\circ$



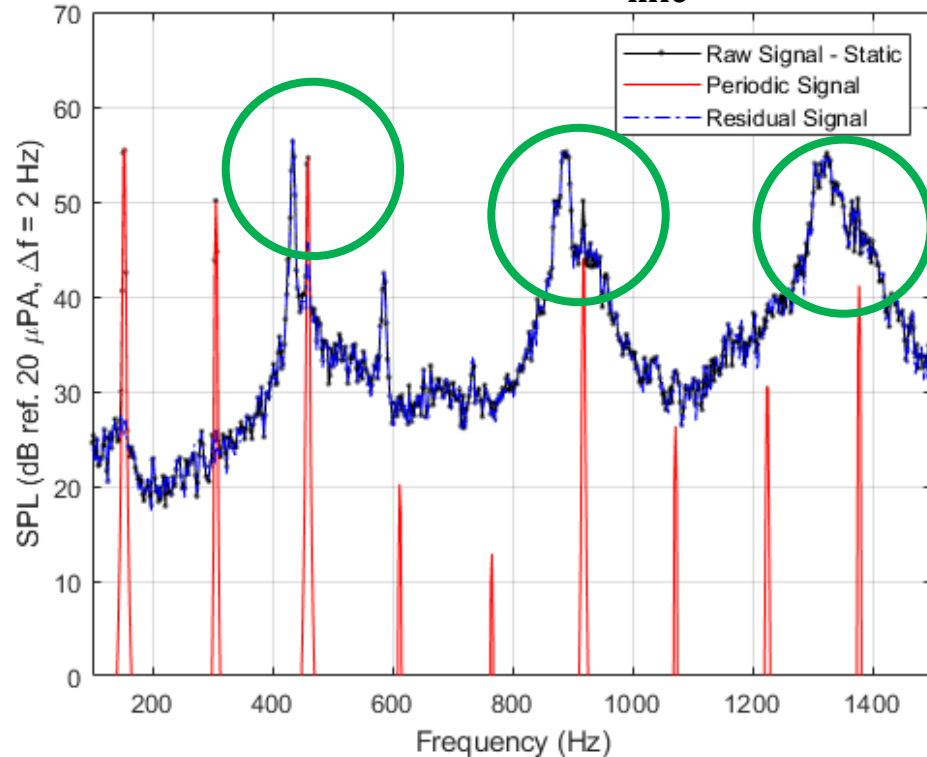
- With no background flow on (static), the spectrum shows additional tonal content below the low harmonic BPFs.
- This is consistent at multiple rotation rates and could be due to inlet separation.
- A background flow of approximately 5 m/s was turned on in the SHAC.
- This removed the additional low frequency tones, as well as “splitting/spreading” behavior.
- The freestream may be helping reattach the flow

Why have the background free stream flow on?



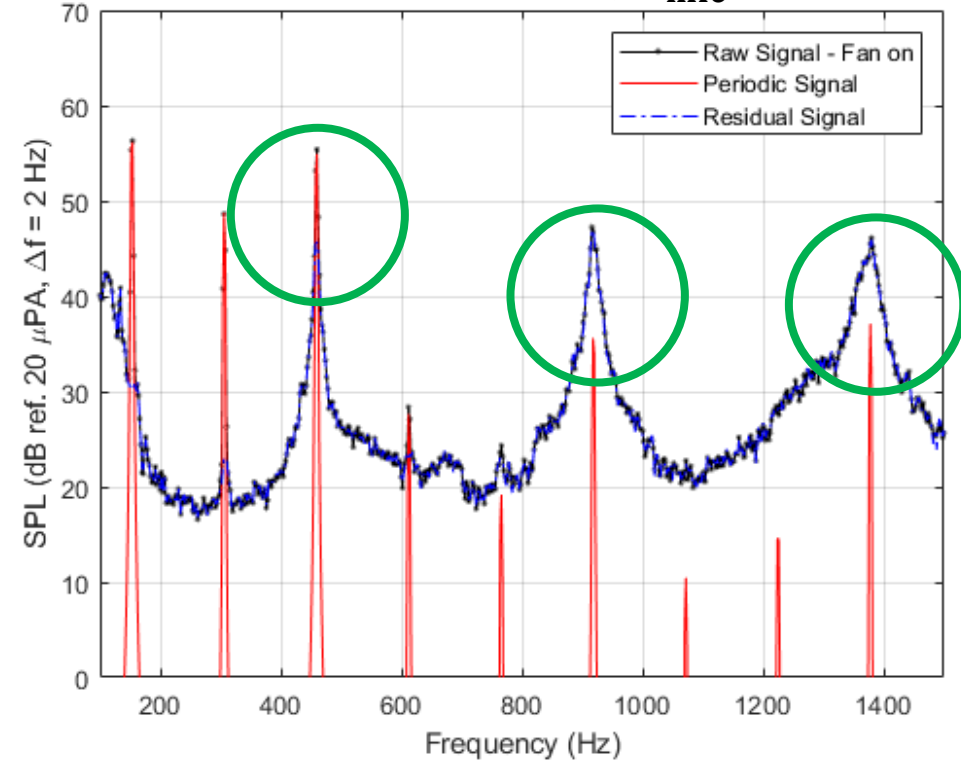
Hardwall Duct US

Static, 9198 RPM, $\theta_{mic} = 0^\circ$



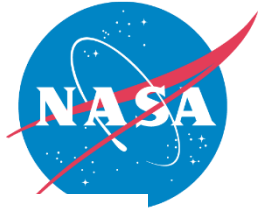
Hardwall Duct US

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Performance Results

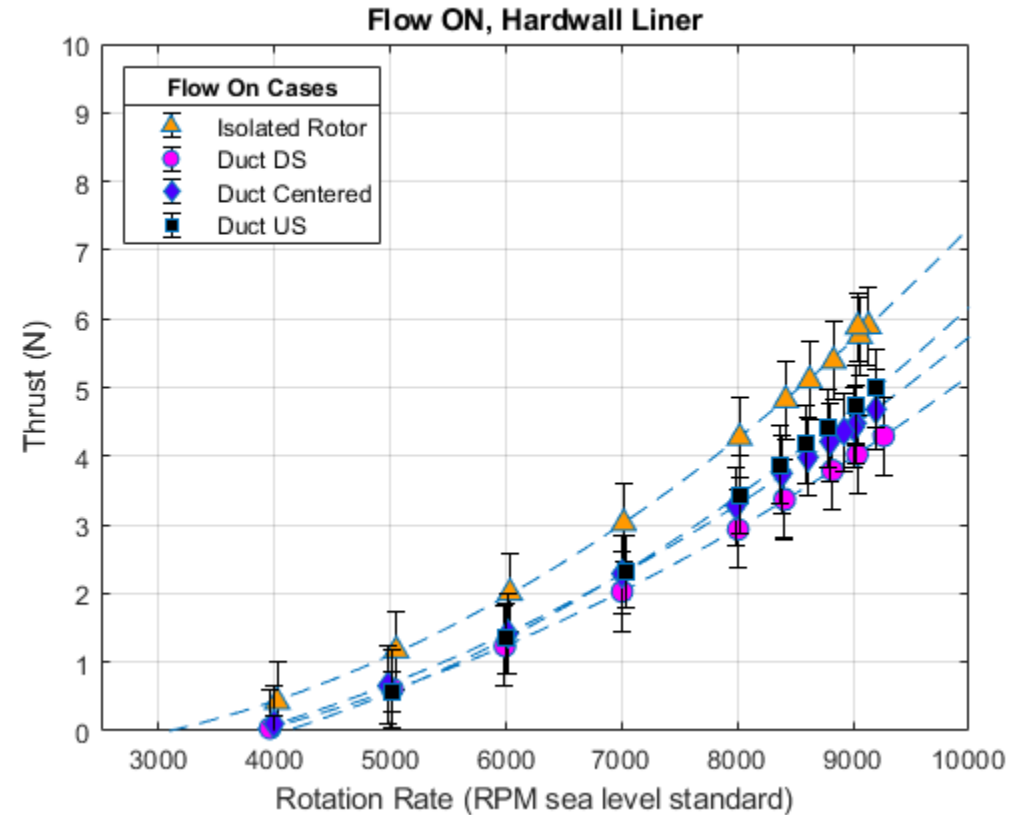


Flow off, 9000 RPM sea level standard

- The **isolated propeller** produces **1.7 lbs** of thrust
- The **ducted propellers** produce **~1.5 lbs** of thrust with the ducts center installed, with **~0.55 lbs** of that being generated by the ducts

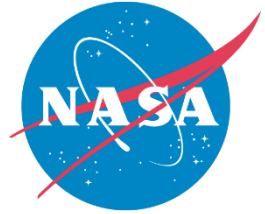
Flow on, 9000 RPM sea level standard

- The **isolated propellers** produces **1.3 lbs** of thrust
- The **ducted propellers** produce **~0.97 lbs** of thrust with the ducts center installed, with **~0.1 lbs** of that being generated by the ducts
- The **net thrust** and **torque increase** when moving the ducts **upstream** for the **flow ON** cases.

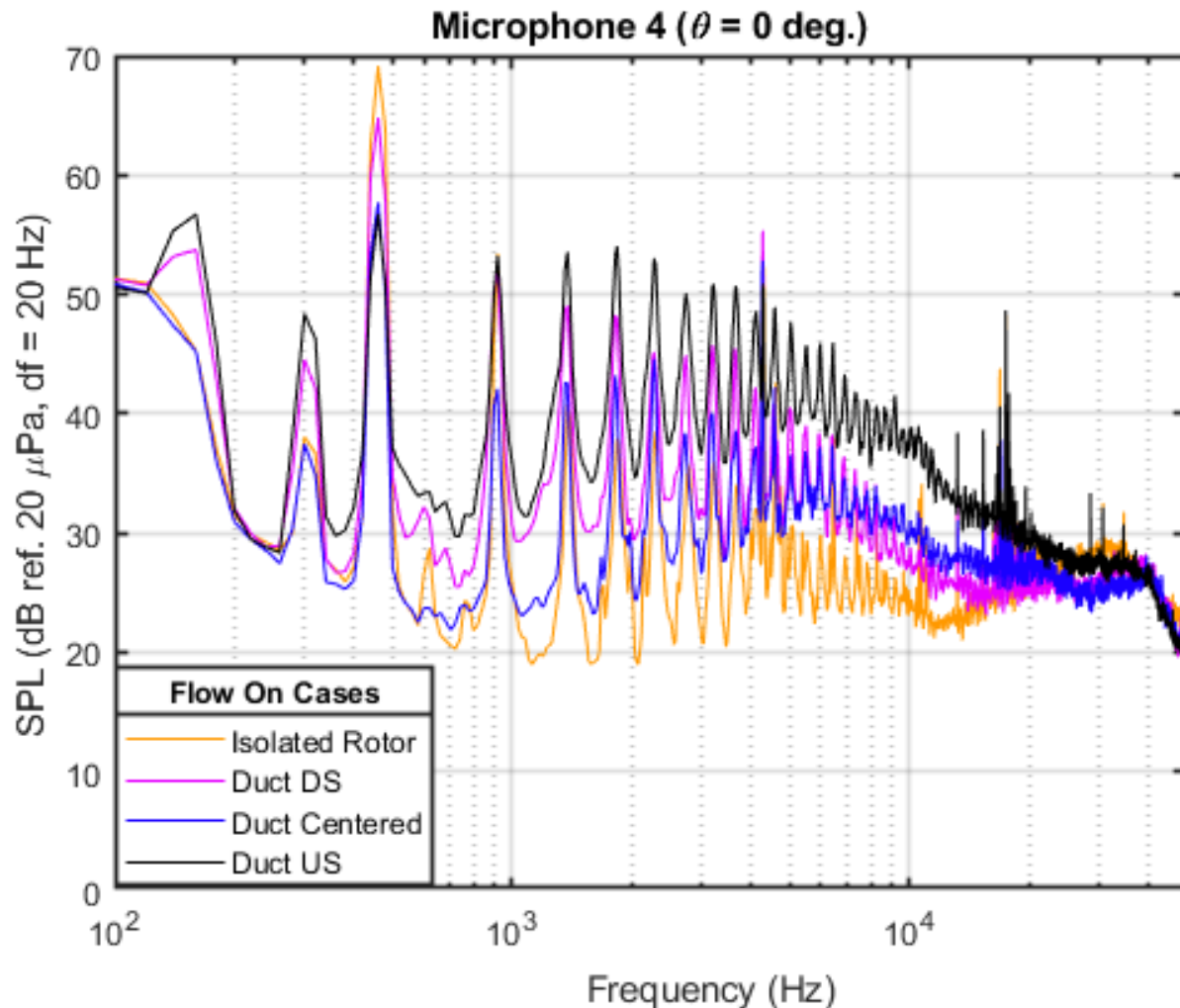


Note: duct is not aerodynamically designed and may not be the most optimized configuration.

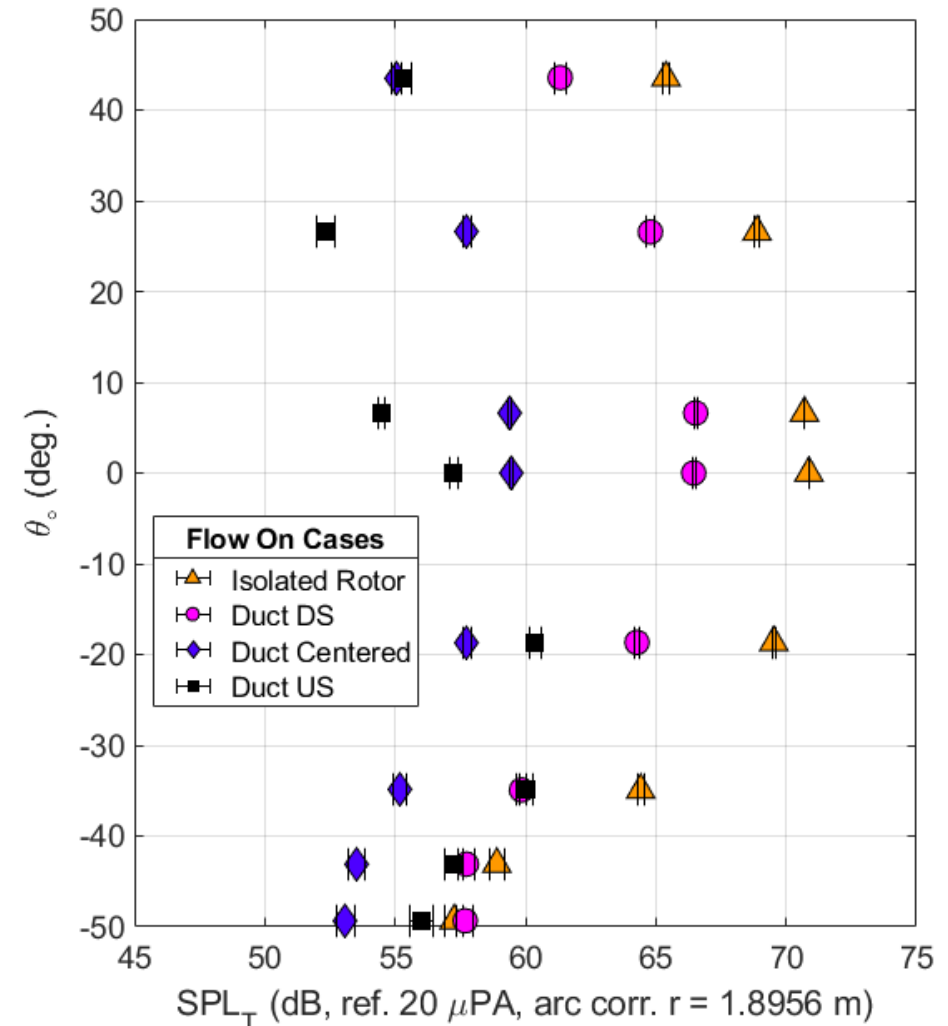
Acoustic Results: Hardwall Duct Flow On



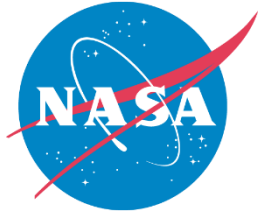
Raw Spectra



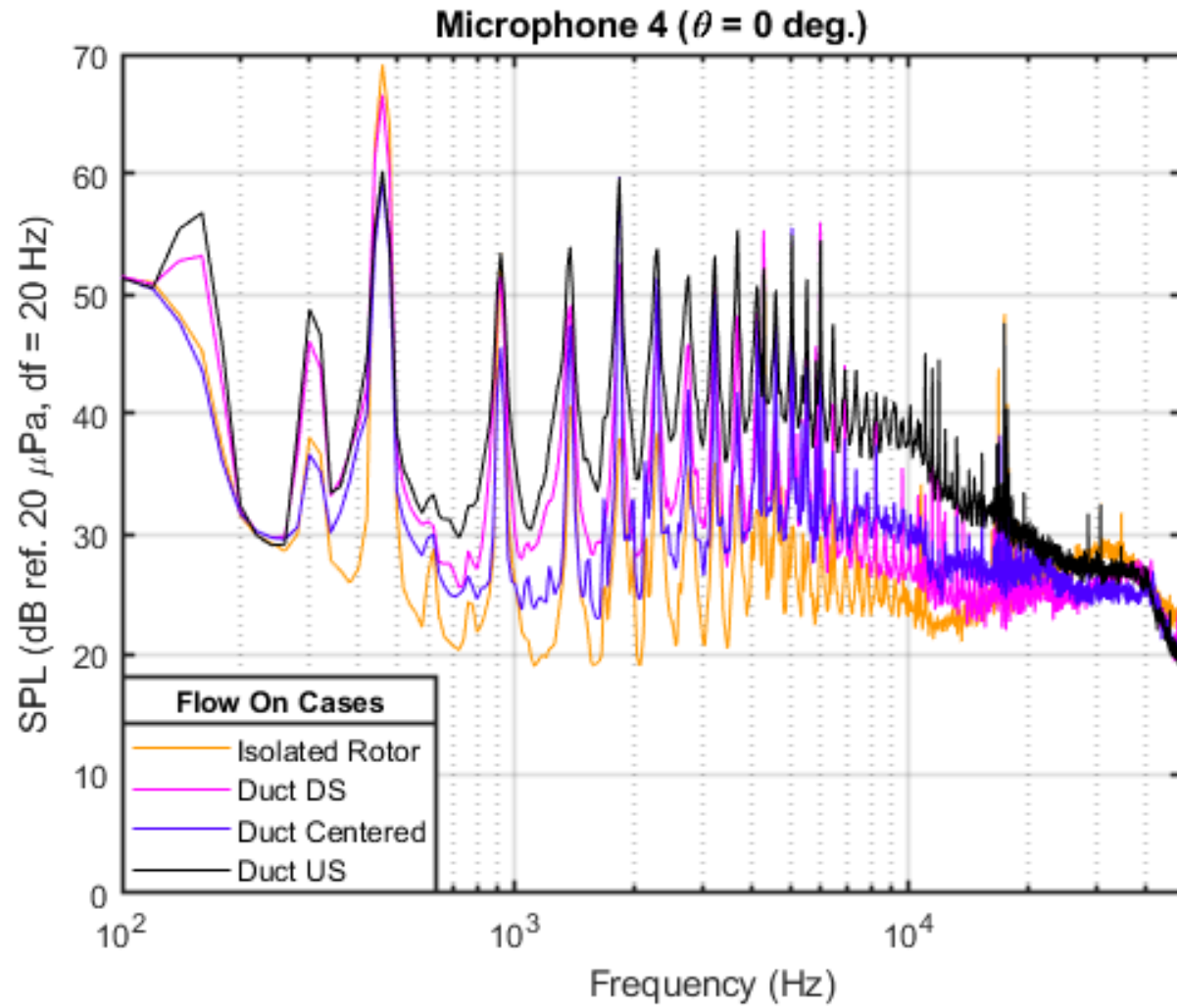
Extracted BPF Tone



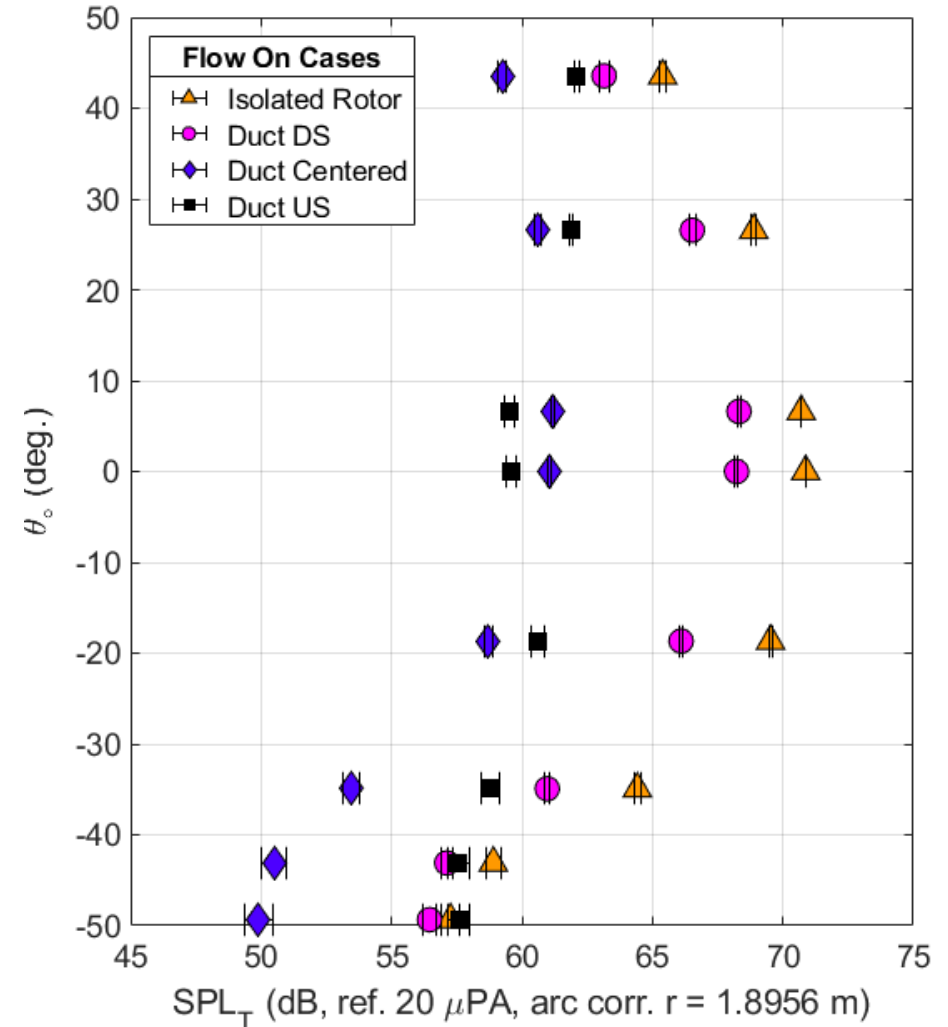
Acoustic Results: L02 Duct Flow On



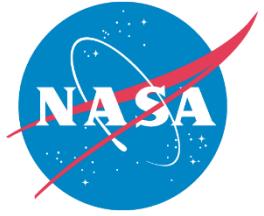
Raw Spectra



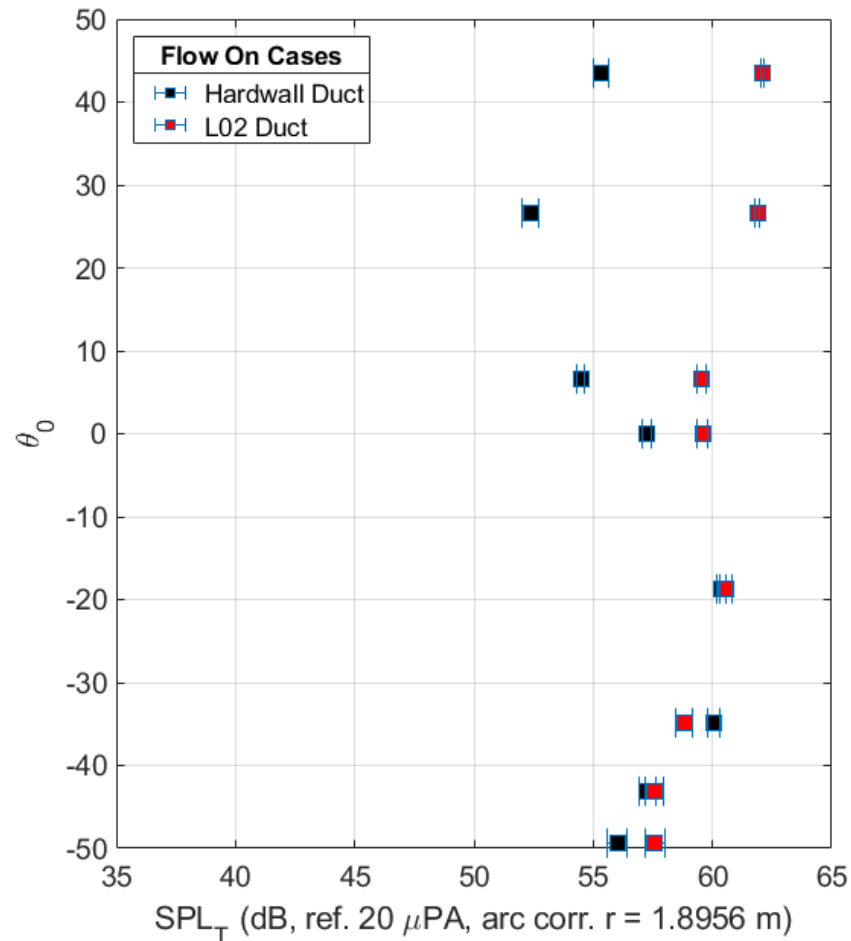
Extracted BPF Tone



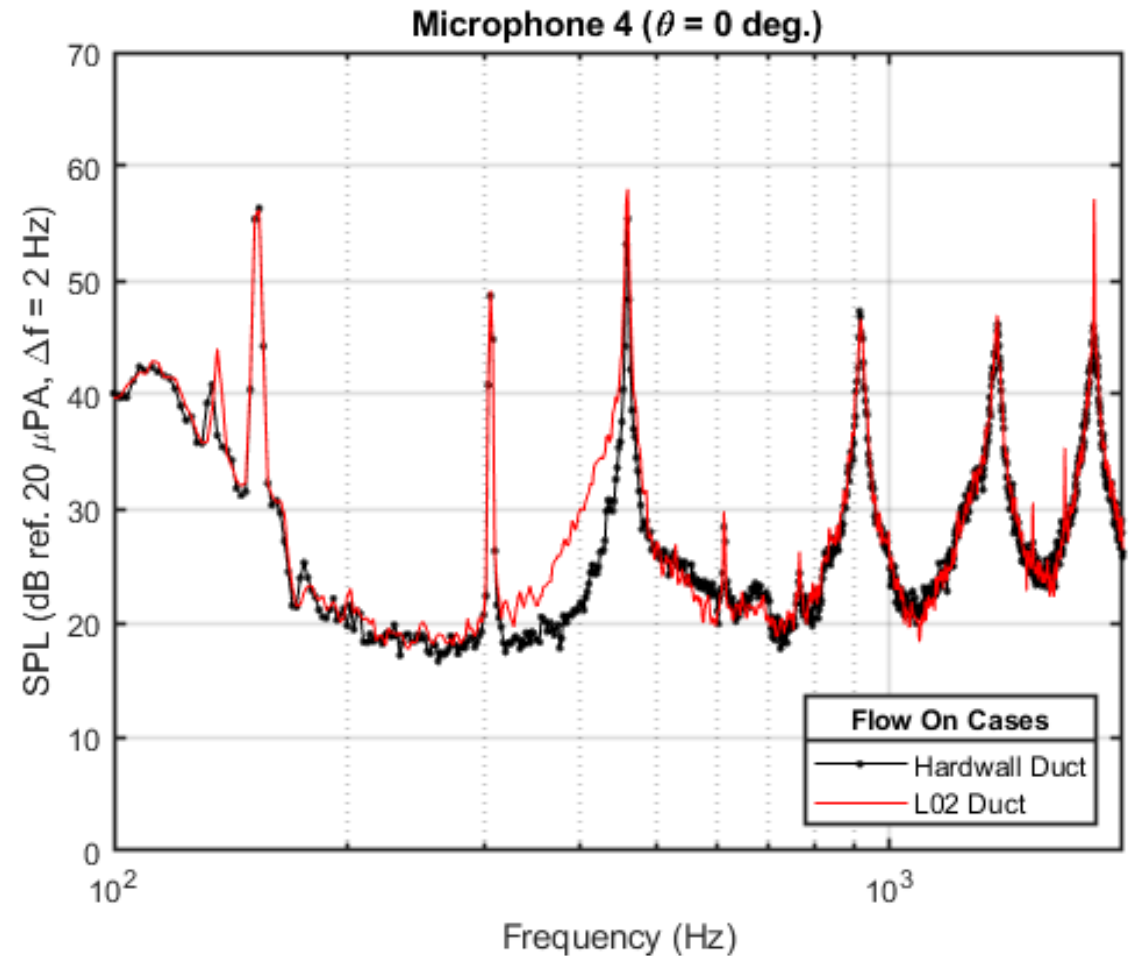
Initial Duct Comparisons – Duct US



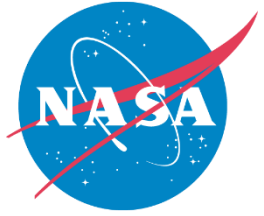
Flow On – Extracted BPF Tone



Flow On – Raw Spectra



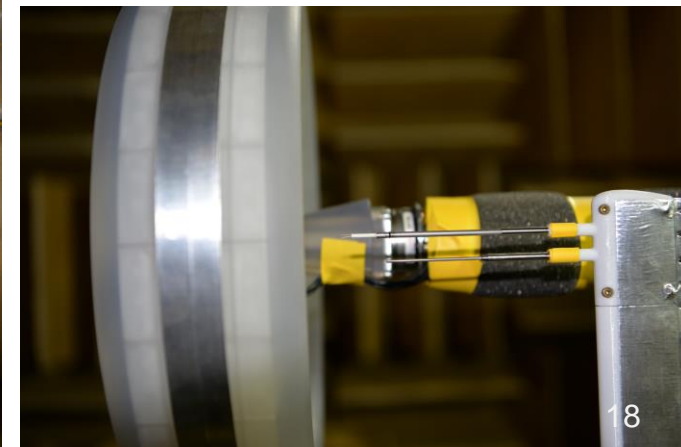
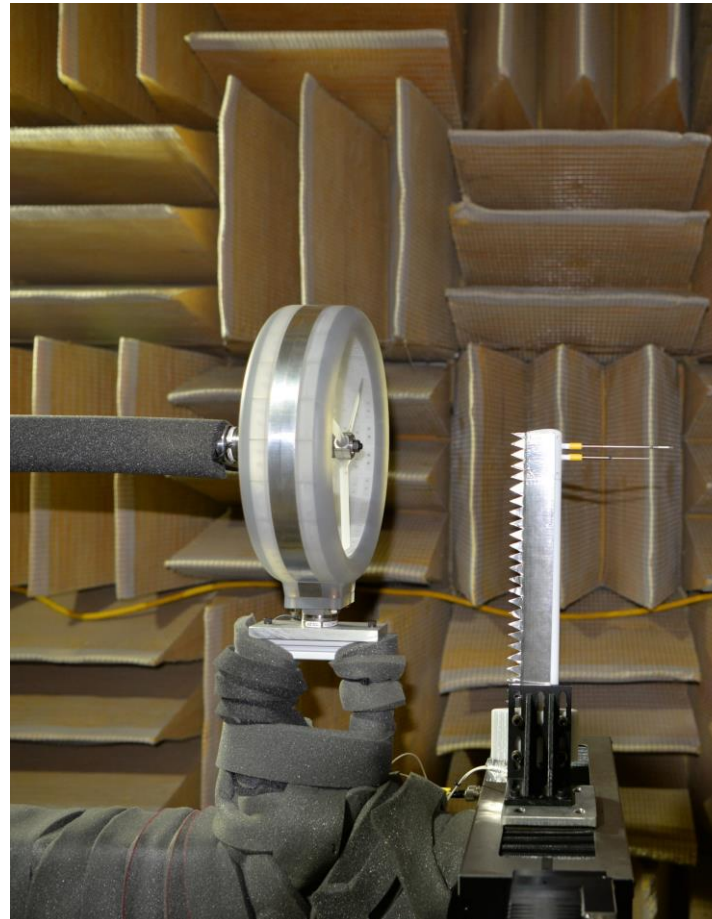
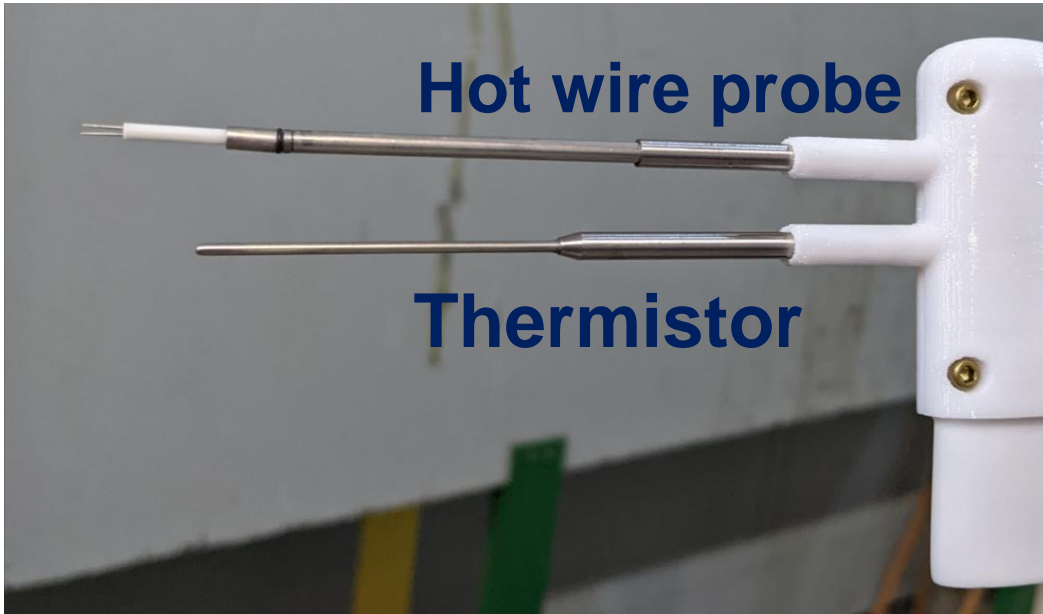
Additional Measurements: Hot Wire Probe



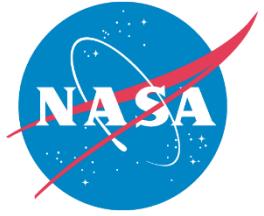
- Motivation
 - To diagnose flow separation near inlet lip
 - To get a better sense of hydrodynamics responsible for large increase in broadband noise
- Two surveys
 - Freestream hot wire survey
 - Wake survey
- Two probes
 - Mini CTA Anemometer 54T42 with 55P16 hot wire probe
 - 90P10 thermistor

Free Stream Survey

Wake Survey

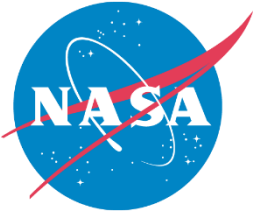


Conclusions



- Experiment data were obtained for various ducted propeller configurations
- Future work
 - Process hot wire data
 - Process acoustic and performance data and identify trends in BPF
- Things to Consider
 - When the duct is installed, the system produces less thrust at the same RPM than when the propeller is isolated. Therefore, a better comparison may be made at a low isolated rotor RPM case
 - When comparing the tonal content of duct cases, it may be best match mechanical rotation rate, as opposed to the corrected sea level standard rotation rate, because the liner is tuned to a certain frequency

Acknowledgments



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LSAWT Crew: John Swartzbaugh, Stan Mason, Jeff Collins, Bryan Lamb,
Mick Hodgins

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757-864-7912

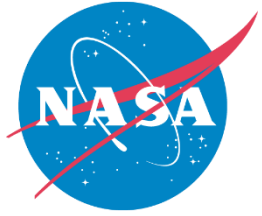
Thank you, any questions?



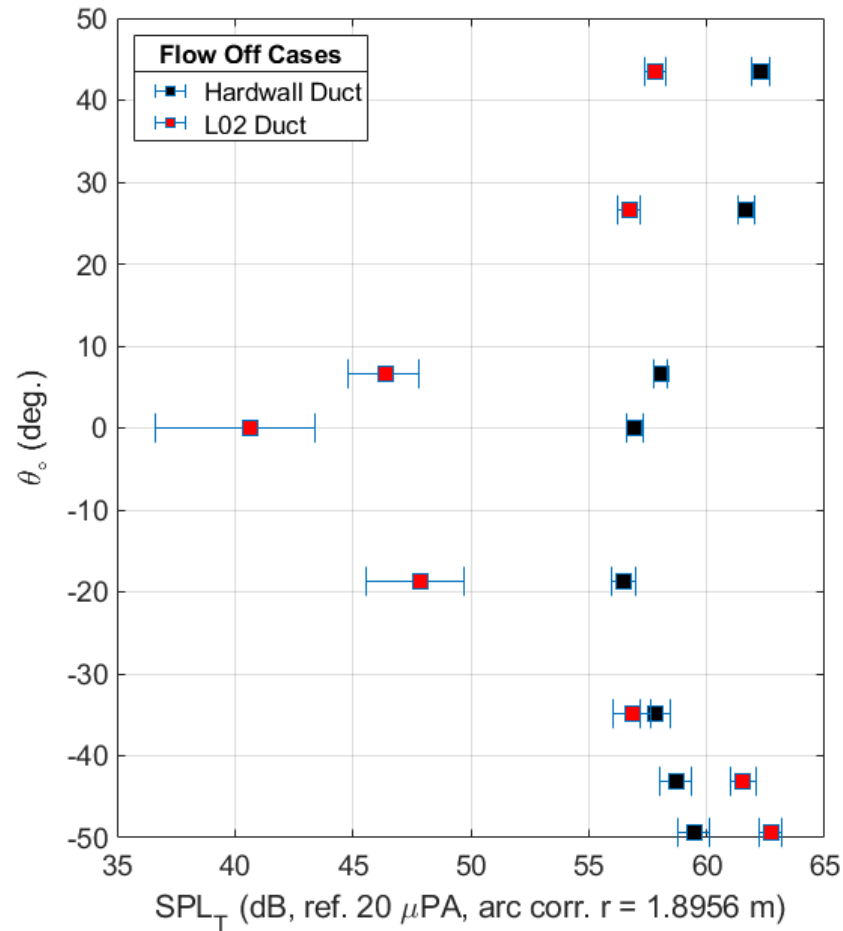
Extra Slides



Initial Duct Comparisons – Duct US



Flow Off – Extracted BPF Tone



Flow Off – Raw Spectra

