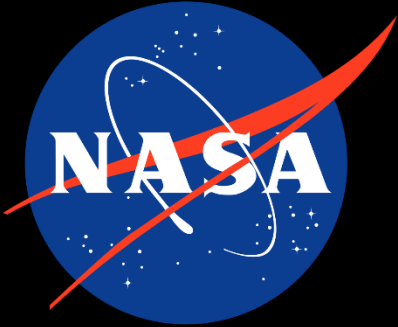




EXPLORE MOON *to* MARS

Process Development and Hot-fire Testing of Additively Manufactured NASA HR-1 for Liquid Rocket Engine Applications

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National Aeronautics and Space Administration (NASA)

***Jacobs Engineering**

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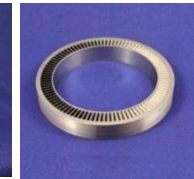
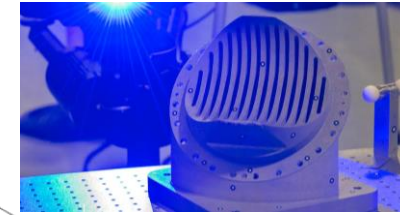
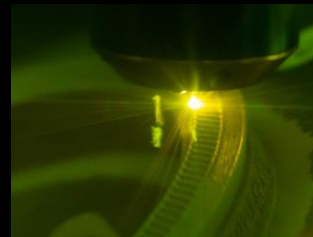
Additive Manufacturing (AM) Development at NASA for Liquid Rocket Engines



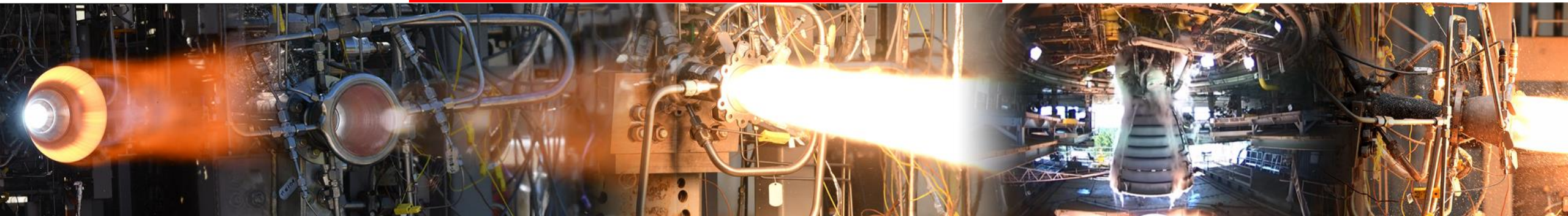
Laser Powder Bed Fusion (L-PBF)
Copper Alloys and Multi-Alloy



Directed
Energy
Deposition

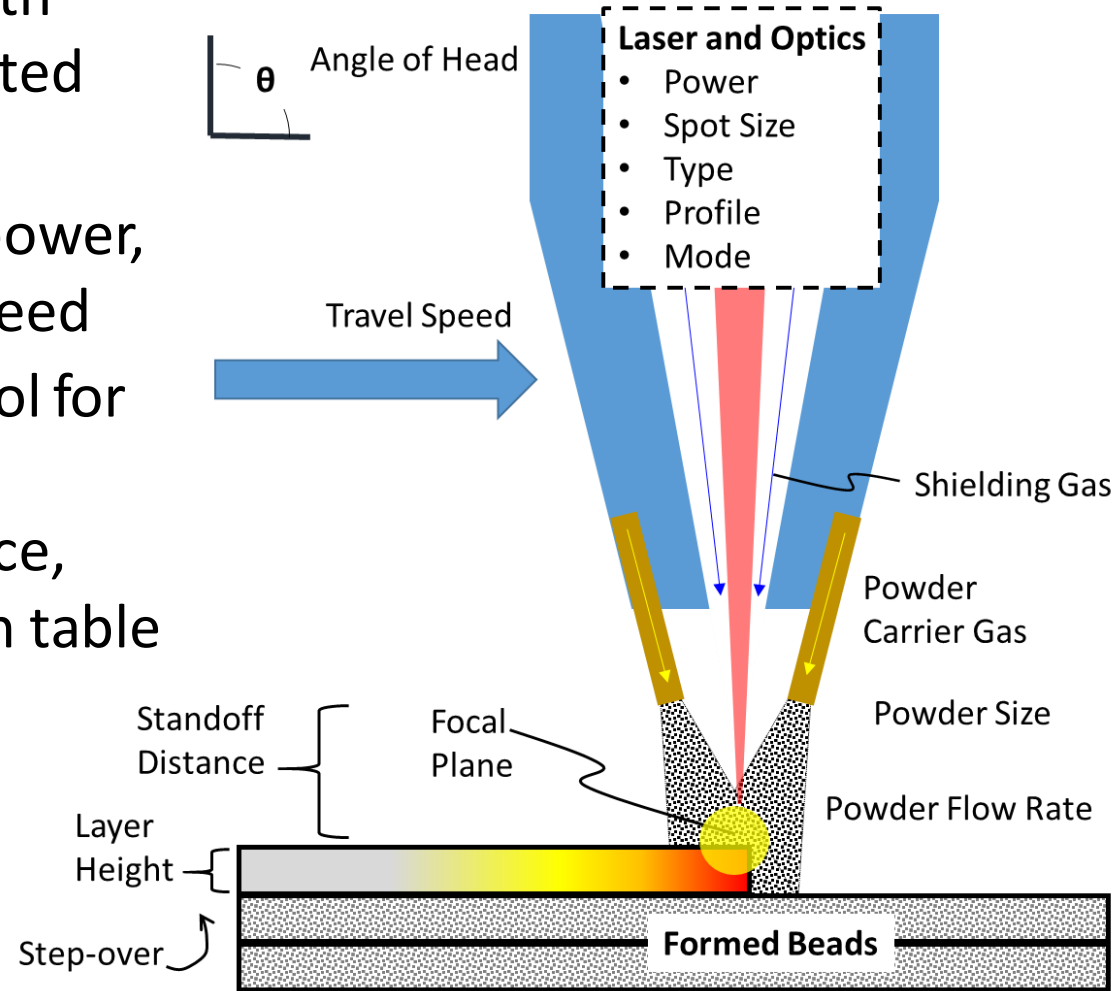


L-PBF of complex components,
new alloy developments for
harsh environment

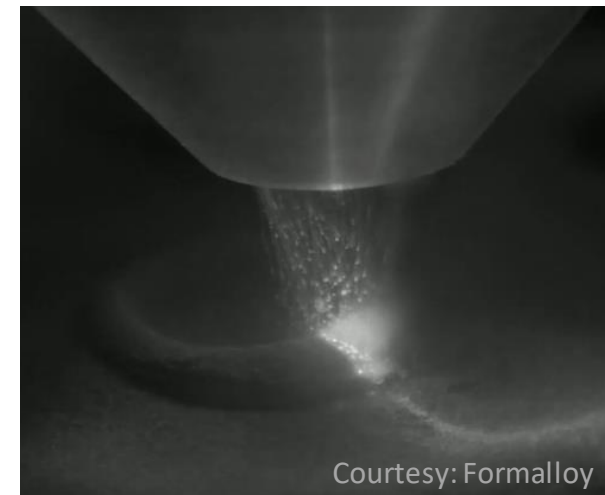


LP-DED Process Overview

- Powder and laser beam path (sometimes optics) integrated into deposition head
- Basic parameters include power, powder feedrate, travel speed
- Additional geometry control for layer height, step over (hatching), standoff distance, angle of head and trunnion table
- Can vary spot size

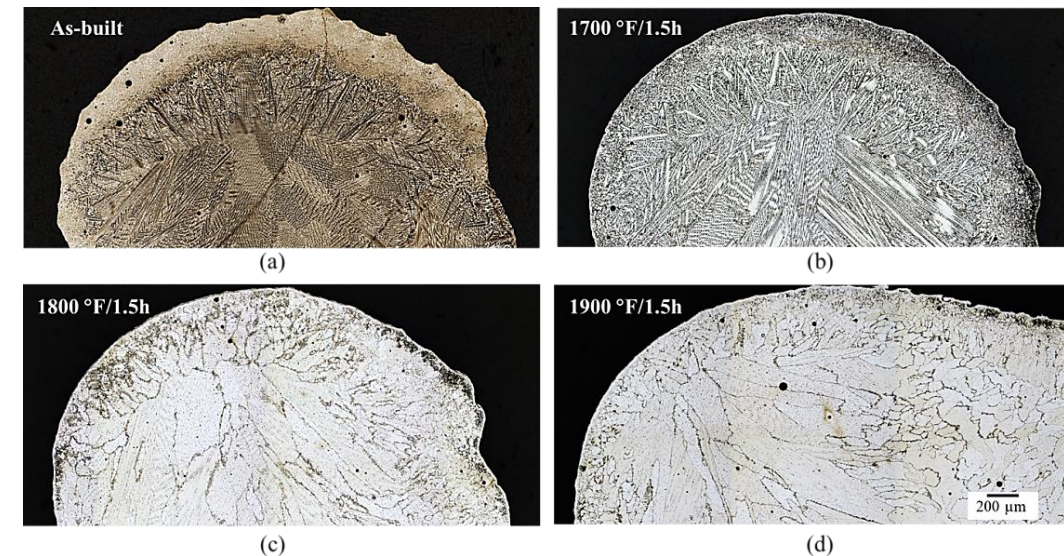
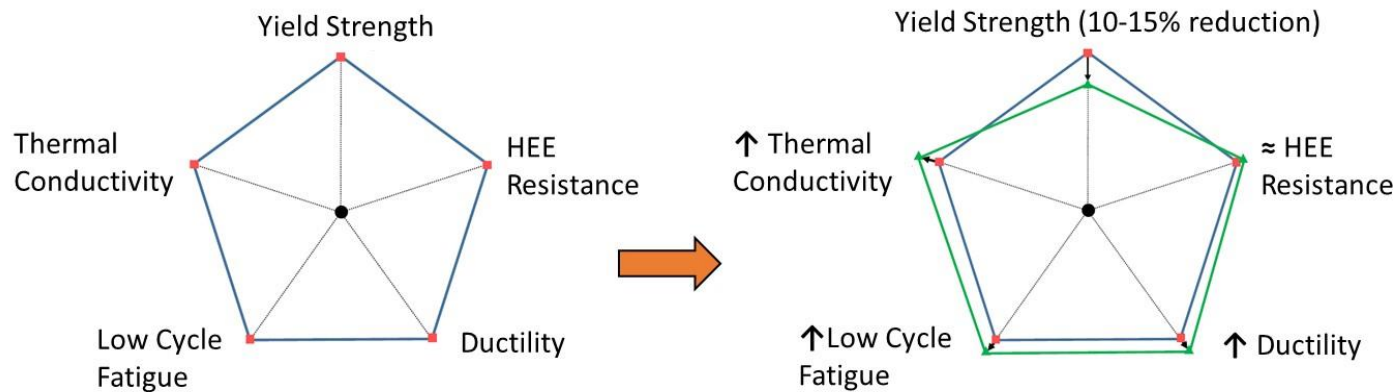


AIAA: Metal Additive Manufacturing for Propulsion Systems, Gradl et al (unreleased)



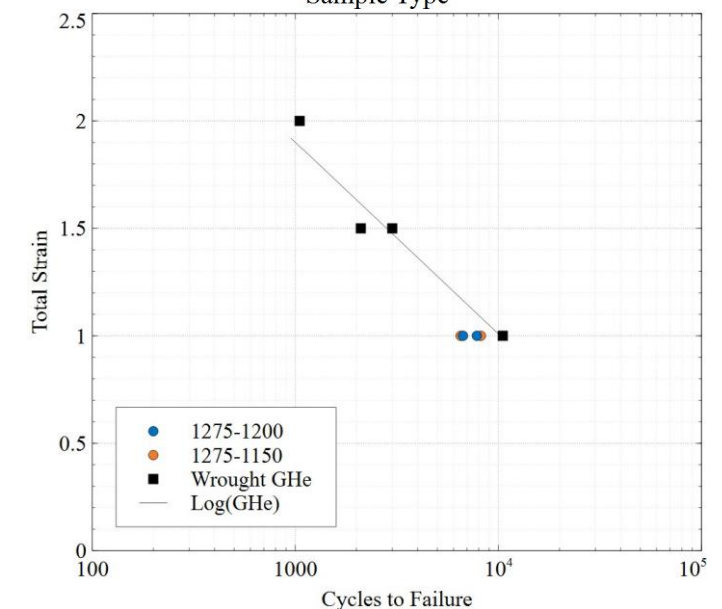
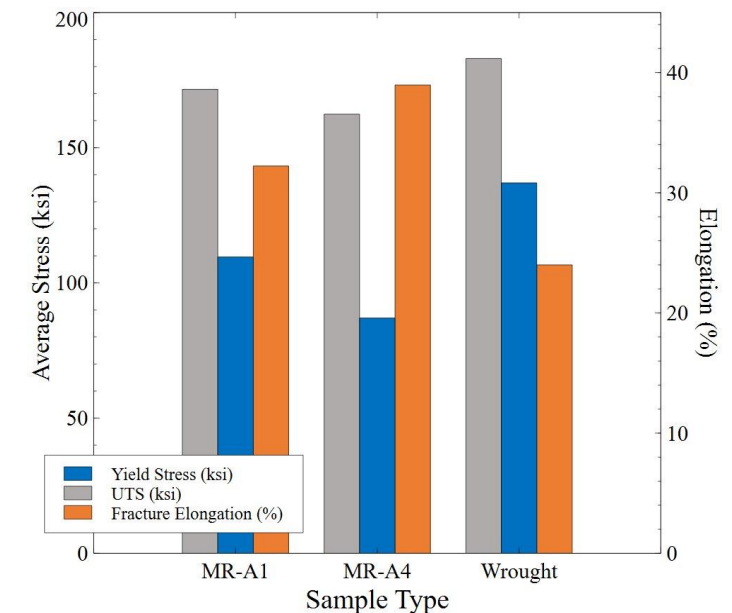
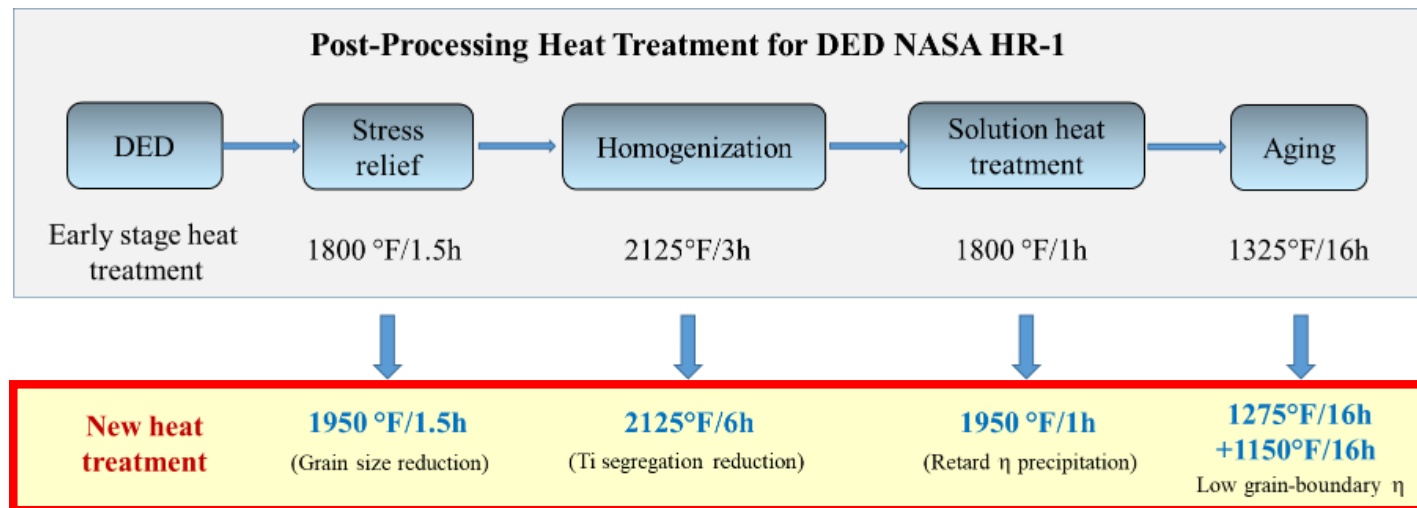
NASA HR-1 Alloy developed for LP-DED Process

- NASA HR-1 is an Fe-Ni-Cr alloy developed for hydrogen environments
- Developed as a derivative of JBK-75 with added W and Co for high strength
- Reformulated for AM DED to reduce Ti-segregation
- Advanced using LP-DED at different deposition rates to allow for variations in wall thickness and deposition time
- Recently advanced heat treatments to mitigate η -phase issues during heat treatment and improve properties

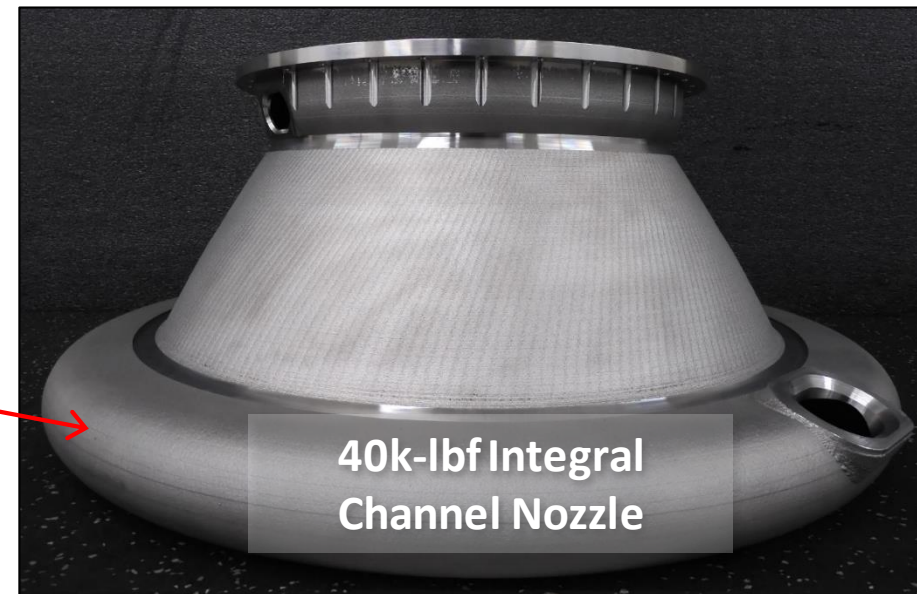


Preliminary NASA HR-1 (Rev 3 Properties)

- A thorough characterization was completed to determine an optimal heat treatment to optimize desired properties
- LP-DED NASA HR-1 exhibits expected trends in strength and ductility



Various Components Fabricated using LP-DED

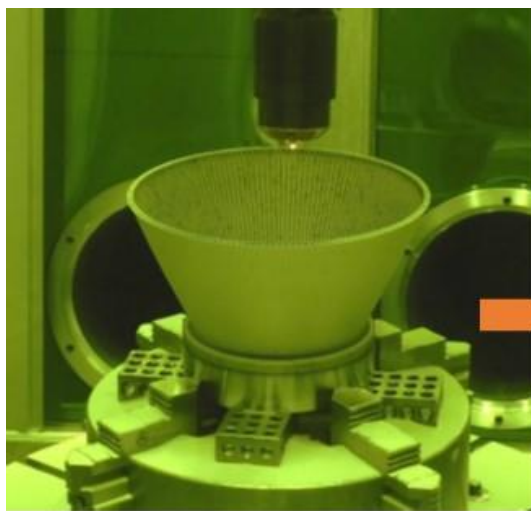


LP-DED Large Scale Nozzle with Fine Features



60" (1.52 m) diameter and >70" (1.78 m) height
90 day deposition

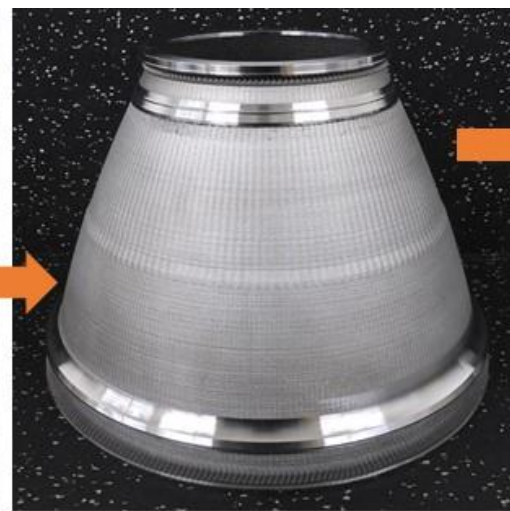
Typical Process flow for LP-DED Nozzles



LP-DED



Post LP-DED



Machining Forward
and Aft Weld Preps



EB Welding
Forward Manifold



GTAW of Aft manifold



Polishing Hotwall



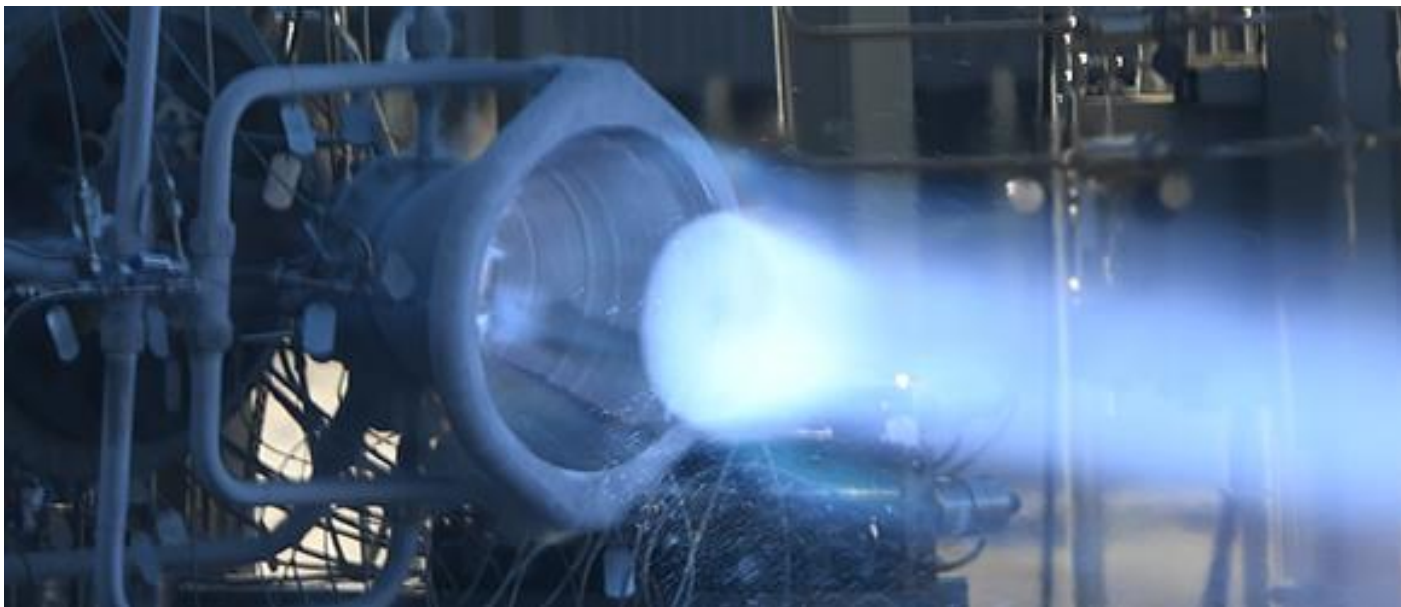
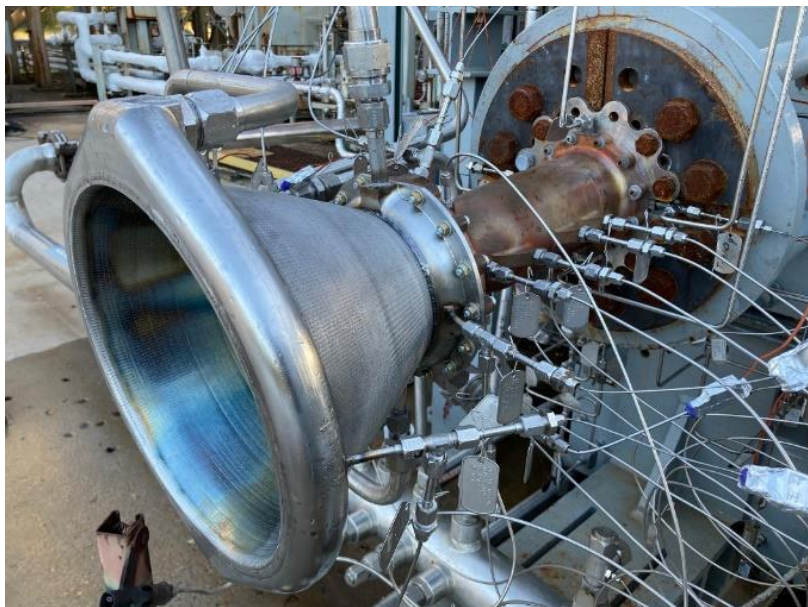
Proof testing



Hot-fire Test Setup for 7,000-lb_f LP-DED Nozzle

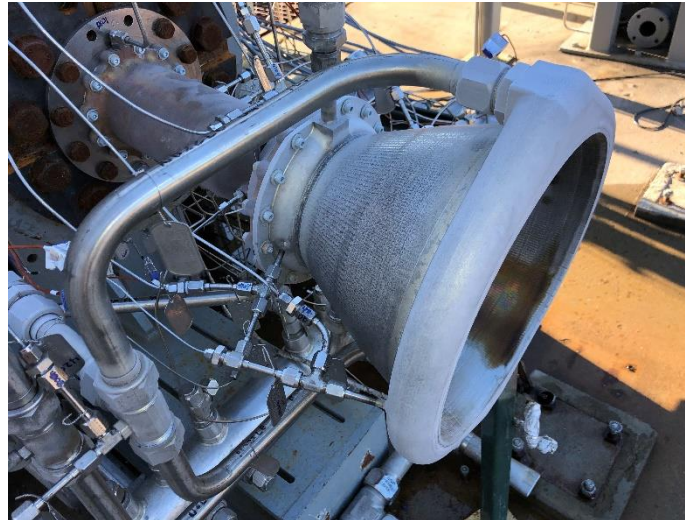


- LOX/Methane with $P_c = 750$ psig and nominal $MR = 3.0$ at MSFC TS115
- Nozzle and chamber cooled with LCH₄
- Integrated with GRCop-42 L-PBF chamber and L-PBF Inconel 625 Impinging injector
- Accumulated a total of 40 tests and 568 seconds
- Low pressure drop in channels
- No issues with nozzle and performance as-expected



Conclusions / Summary

- LP-DED is a viable process to design, fabricate and test large scale components for liquid rocket engines
- LP-DED provides a scale not possible with other metal AM techniques
- NASA HR-1 has been demonstrated as a high strength alloy for channel wall nozzles and other components in hydrogen-environments
- Hot-fire testing accumulated 40 starts and 568 seconds at 7,000-lbf scale
- Various projects to complete hot-fire testing (2.5k to 40k-lbf) using LP-DED
- Data and properties available to commercial and government partners





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Large scale Integrated Channel DED Nozzle Build

