

# In-situ Imaging of Pyrolyzing Aerospace Materials

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&  
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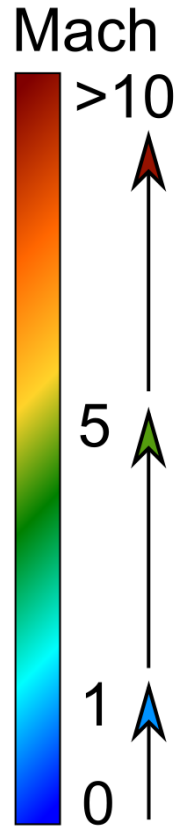


**Grainger College  
of Engineering**  
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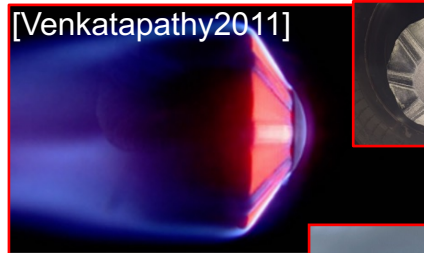


NASA – [80NSSCC22K1192, 80NSSC21K1117]  
DOE ALS – [DE-AC02-05CH11231], DOE SCGSR – [DESC0014664]

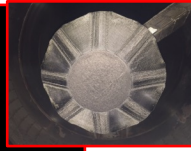
# Aerospace Vehicles



[Venkatapathy2011]



Inter-planetary probes



Meteorites



Defense vehicles



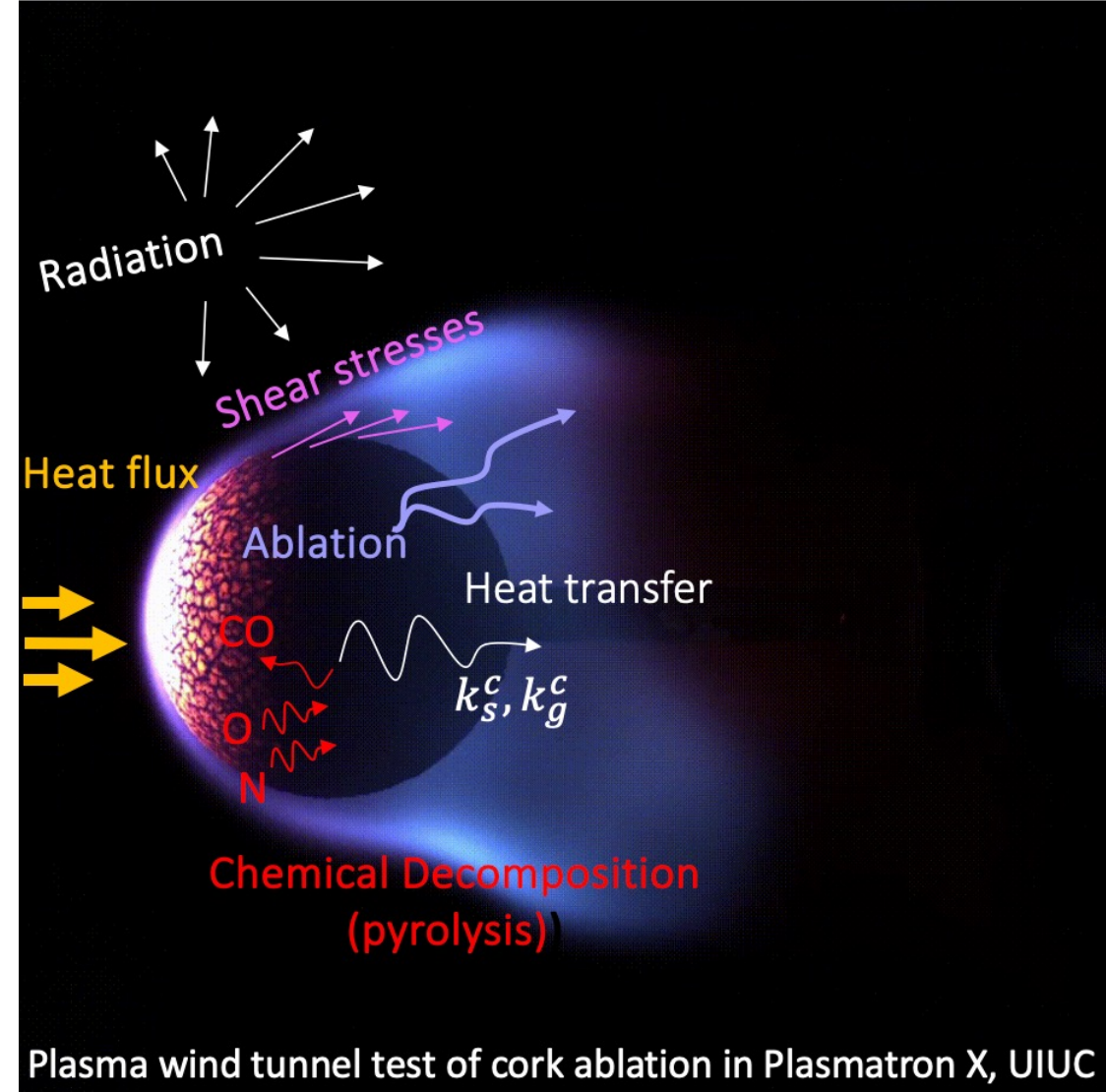
[Walker2008]

Military jets

Reentry parachutes



Commercial airliners

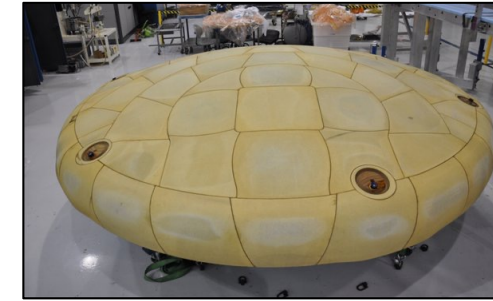


# Overview of materials for hypersonic entry

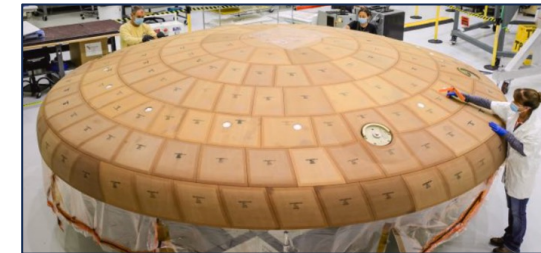
- For atmospheric entry, a sacrificial “ablative” material is used as a heatshield:
  - Carbon or silicon-based composites that endothermically pyrolyze and prevent heat from entering the inner structure.
- NASA has typically used tiled heatshields of PICA or Avcoat, bonded with Room Temperature Vulcanizing silicone (RTV560) gap-filler.
- RTV560:
  - Initially non-porous solid that pyrolyzes, becomes porous and swells as it is heated.
  - As it swells, it can cause transition of flow from laminar to turbulent and increase heating on the heatshield.



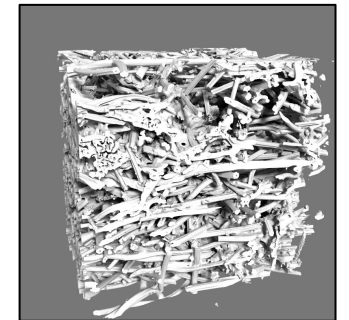
Mars Science Laboratory (MSL) heatshield with tiled PICA bonded by RTV560 [1].



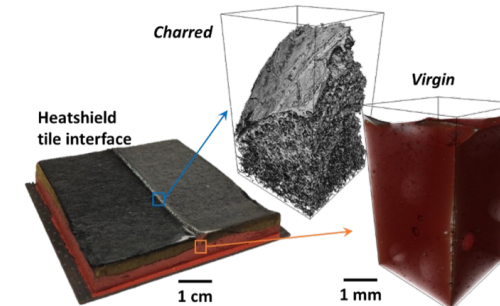
SpaceX Dragon heatshield with tiled C-PICA [Credits: SpaceX].



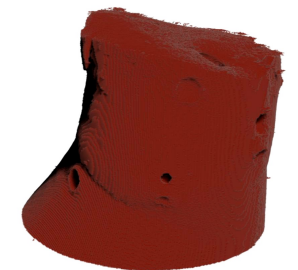
Artemis I heatshield made of Avcoat with RTV560 gap filler [Credits: NASA].



Tomography of carbon fibers in PICA [Credits: NASA].



Tiled heatshield showing virgin and charred RTV. [2]



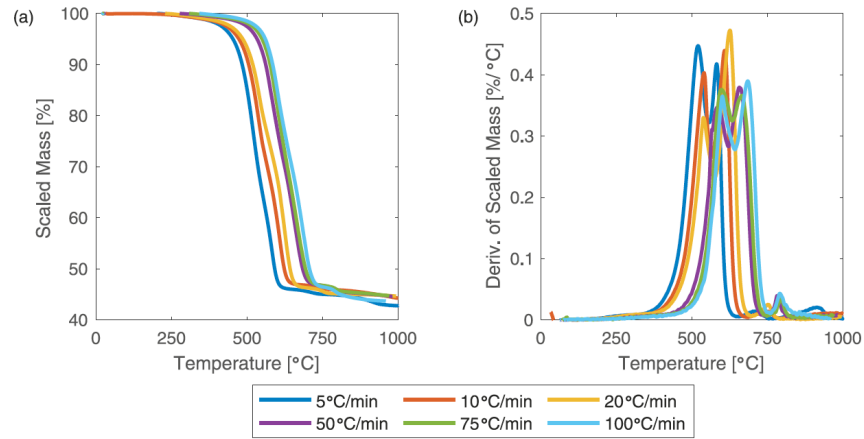
In-situ tomography of RTV showing swelling and shrinking.



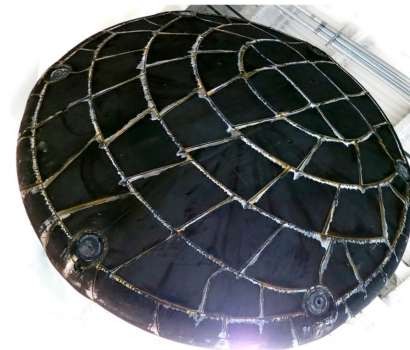
# RTV 560 Pyrolysis and Intumescence

## ➤ RTV560:

- Silicone elastomer with  $\text{Fe}_2\text{O}_3$  grains and diatomaceous Earth,
- Loses a significant amount of mass and changes in porosity and volume drastically [2].



Courtesy of Space X

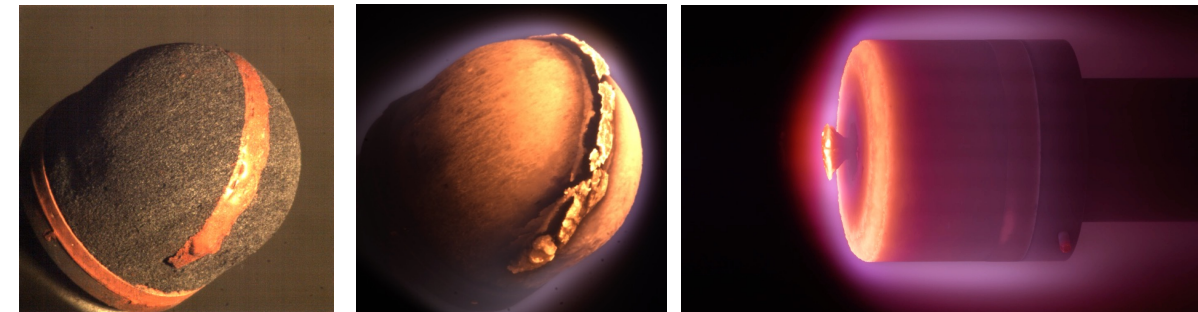
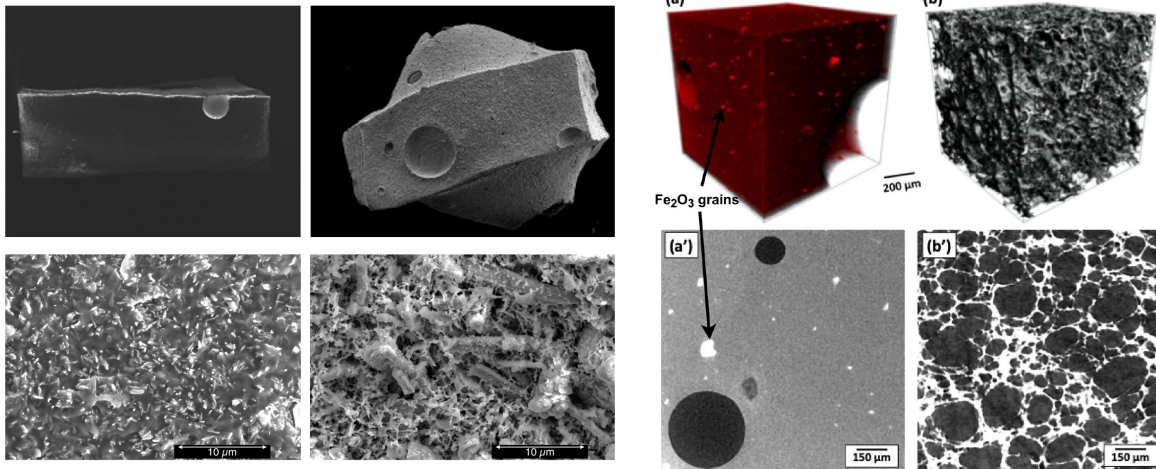


Post-flight Dragon

Courtesy of NASA Ames



RTV swelling in TPS [3].



RTV tested at increasing heat fluxes.

# RTV 560 Pyrolysis and Intumescence

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*Courtesy of Space X*

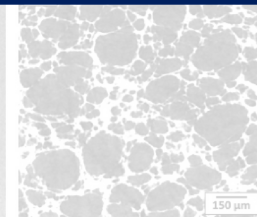
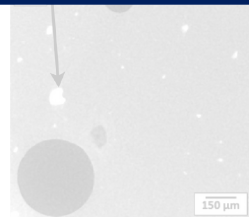
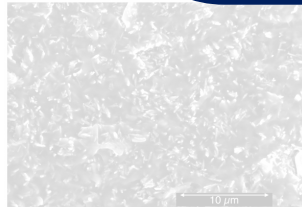
*Courtesy of NASA Ames*

## Goal of current work:

- Determine microstructure change of RTV as function of temperature using MicroCT at the Advanced Light Source,
- Obtain quantitative estimates for macrostructural change,
- Compute properties such as permeability, thermal conductivity, etc. at different temperatures.

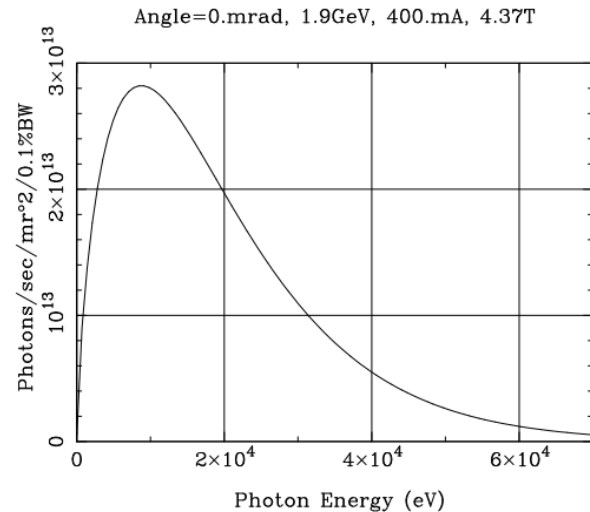


S [3].

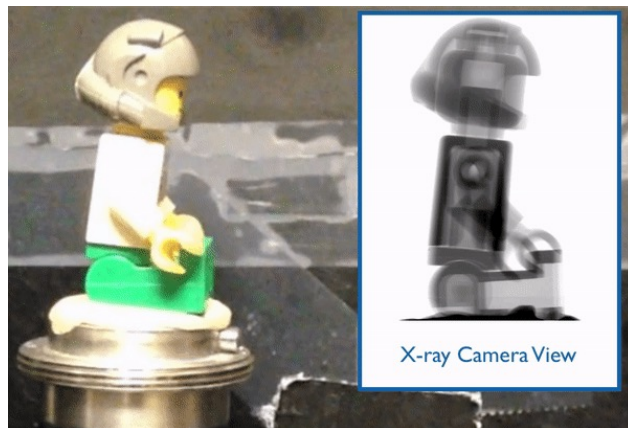


RTV tested at increasing heat fluxes.

# Tomography at ALS, Beamline 8.3.2



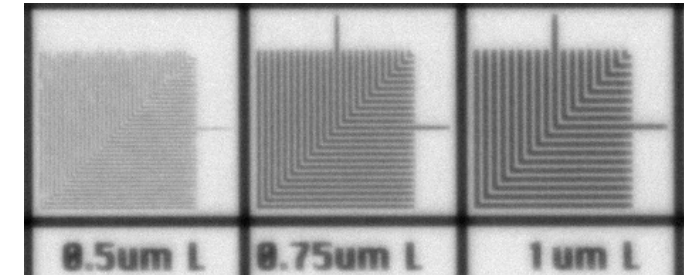
X-ray flux at different energies at Beamline 8.3.2.



Synchrotron  $\mu$ -CT of Astronaut, Courtesy of Dula Parkinson

	<u>PCO Edge</u>	<u>(2560x2160)</u> <u>[Optique Peter]</u>
Lens	pixels size (um)	FOV (mm)
20x	- [0.33]	- [0.8]
10x	0.65 [0.69]	1.7 [1.7]
5x [4x]	1.3 [1.72]	3.3 [4.4]
2x	3.25 [3.44]	8.3 [8.8]
1x	6.5 [-]	16.6 [-]

Different lens options available for tomography.



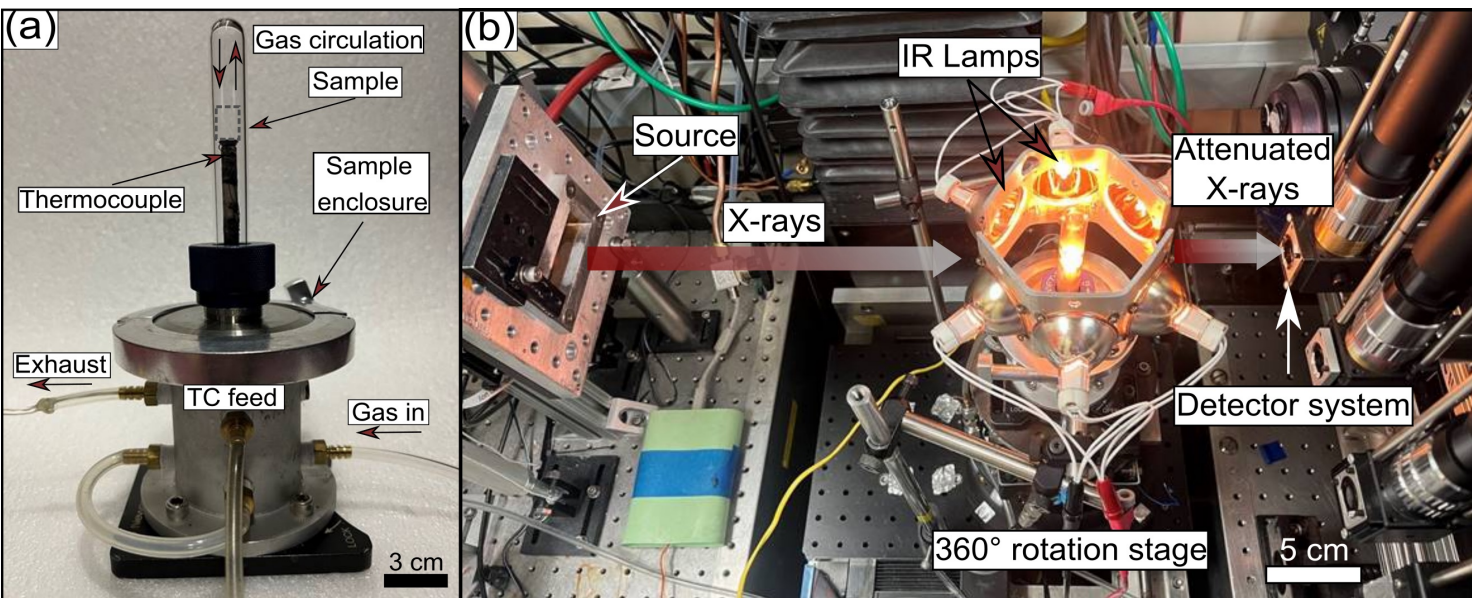
Finest resolution scan (~0.5 micron) with 20x lens.

X-ray system	Resolution [ $\mu$ m/px]	FOV, $\varnothing$ [mm]	Time/tomography	Capabilities
Xradia Bio-MCT	0.9 – 10	1.5 – 10	> 2 hours	<i>ex-situ</i>
ALS 8.3.2 Beamline	0.6 – 10	1.7 – 27	30 s – 30 min	<i>in situ</i>

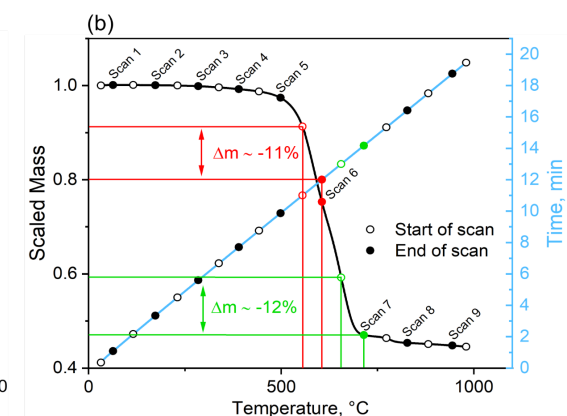
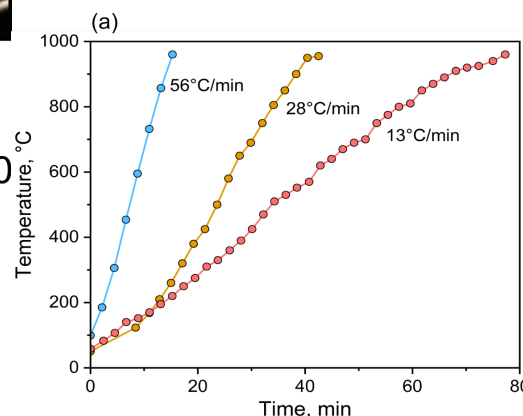
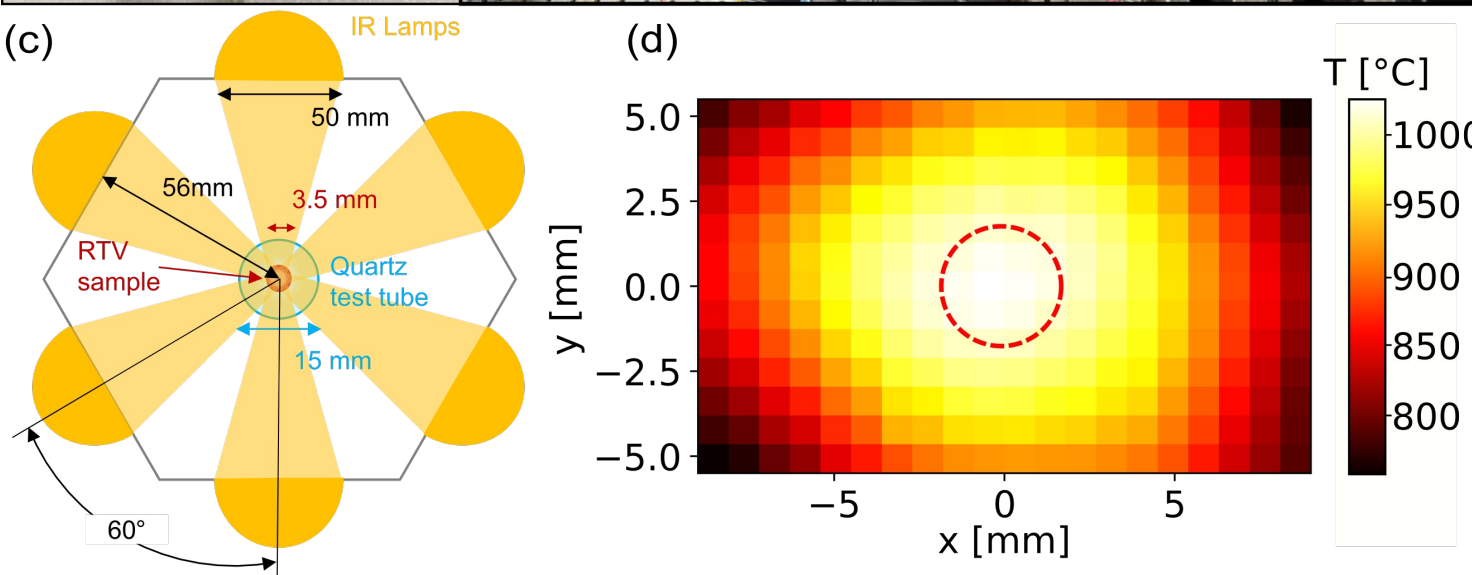


# Pyrolysis setup for RTV [4,5]

- RTV samples are heated by IR lamps and scans are collected every 90 seconds.

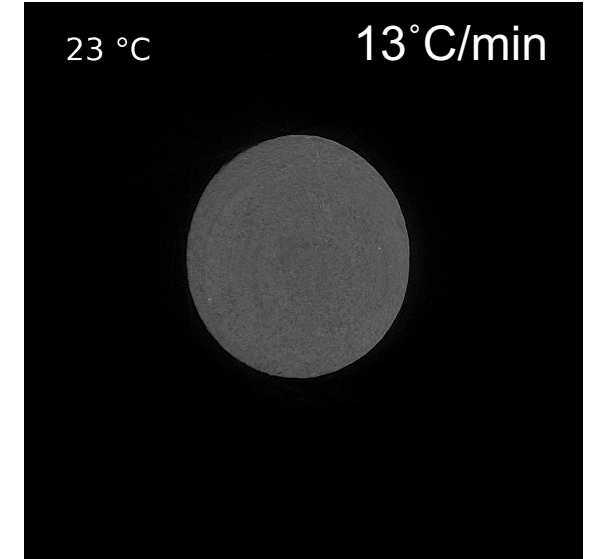
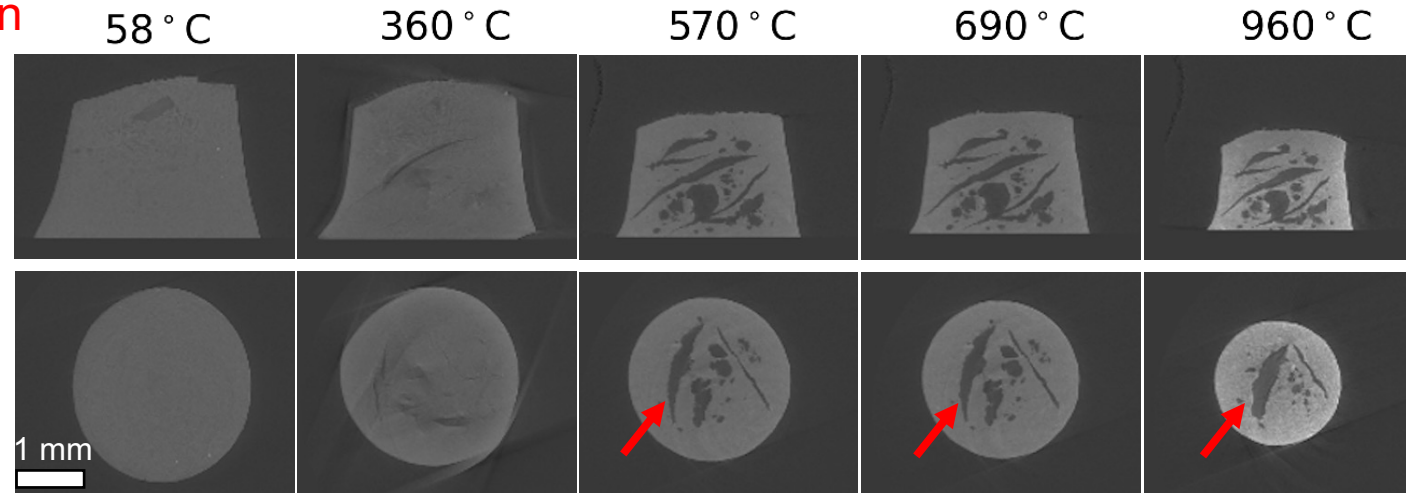


- (a) Environmental chamber  
(b) Inside  $\mu$ -CT hutch at ALS  
(c) Lamp schematic  
(d) Spatial temperature variation

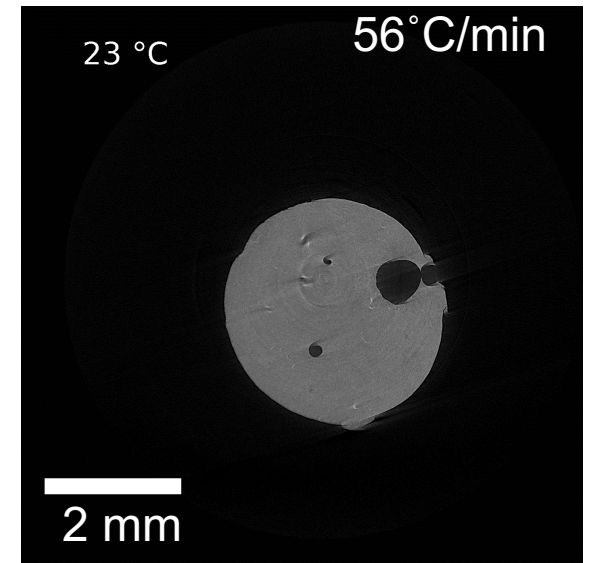
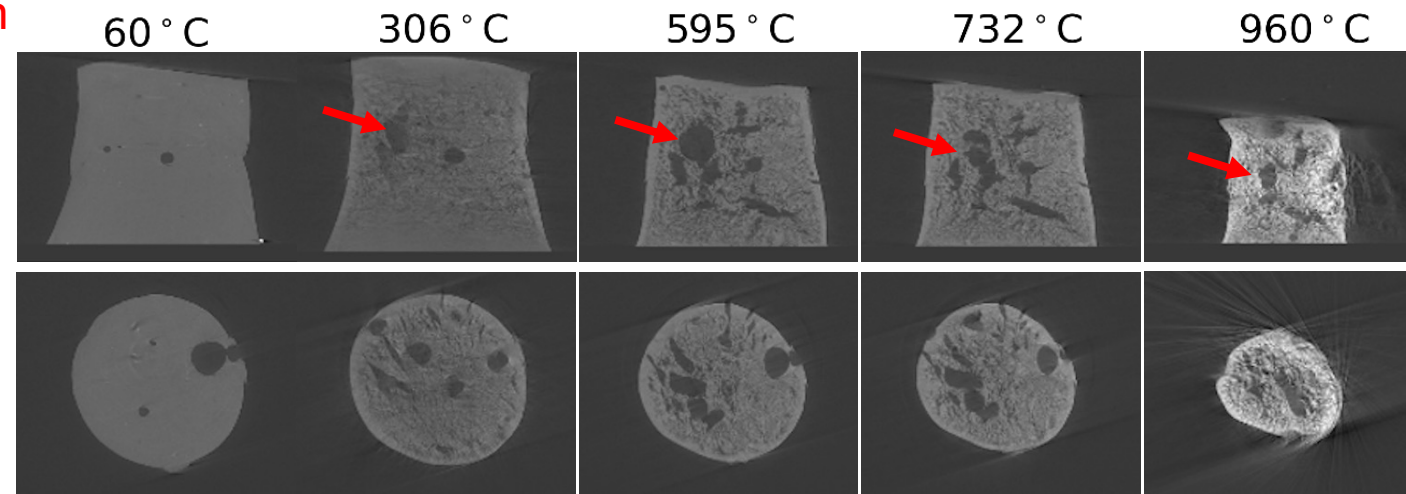


# RTV Pyrolysis

13°C/min

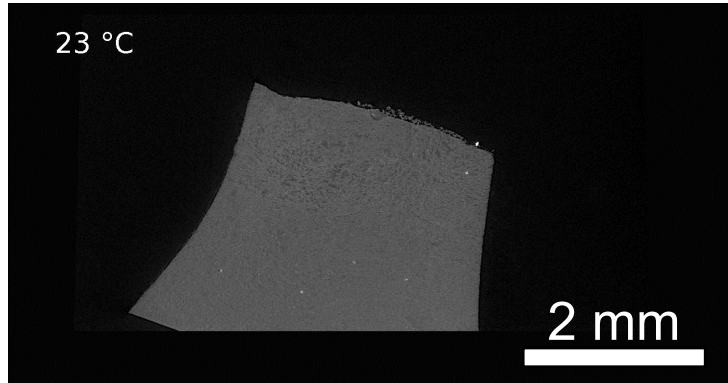


56°C/min

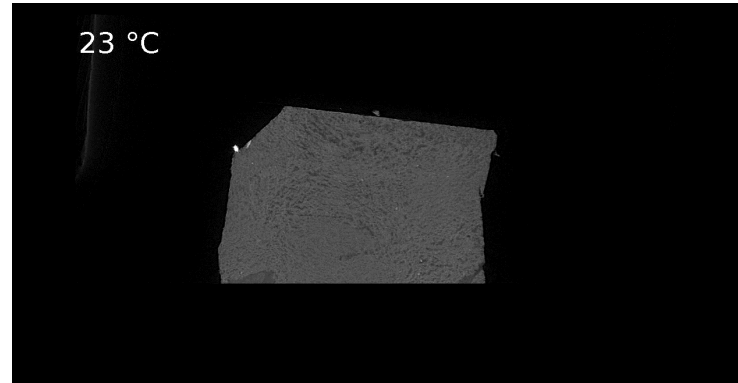




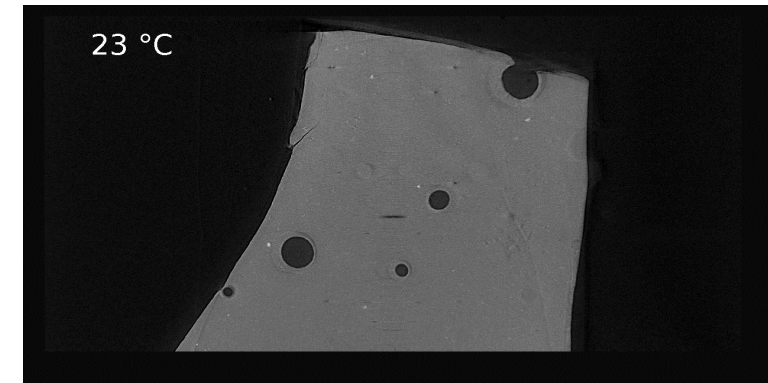
# Quantitative estimates for morphological evolution



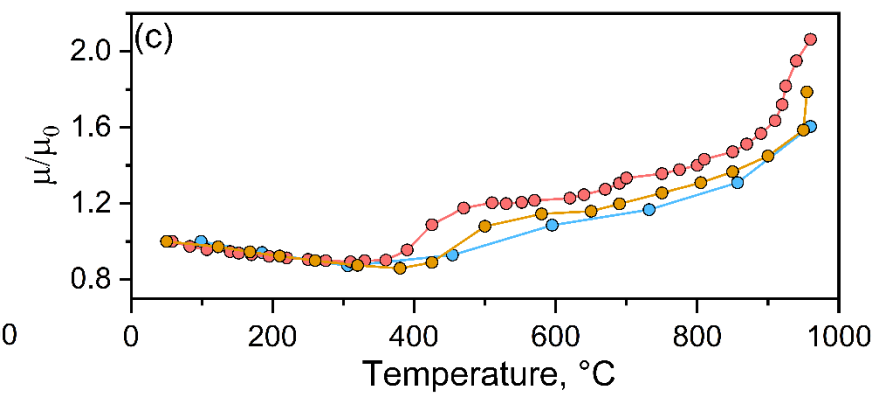
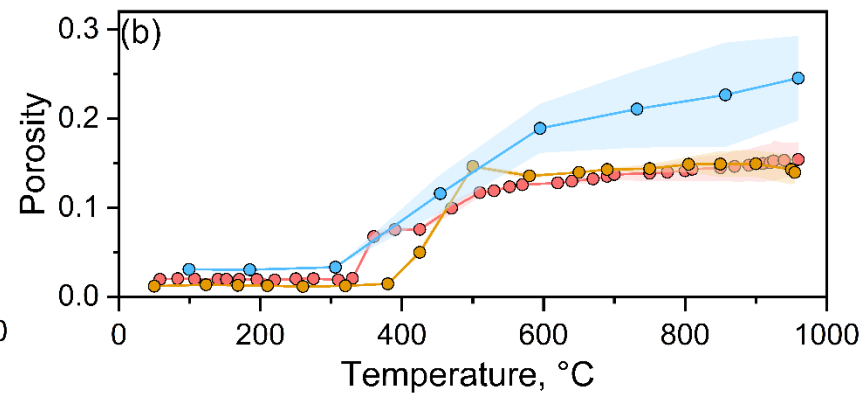
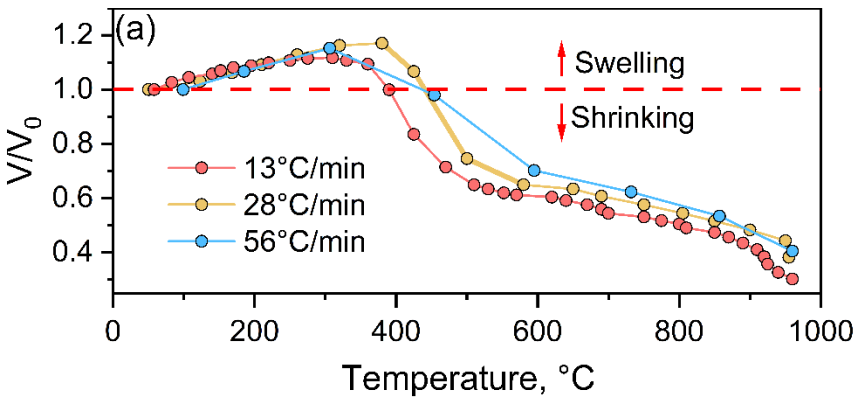
13 °C/min



28 °C/min

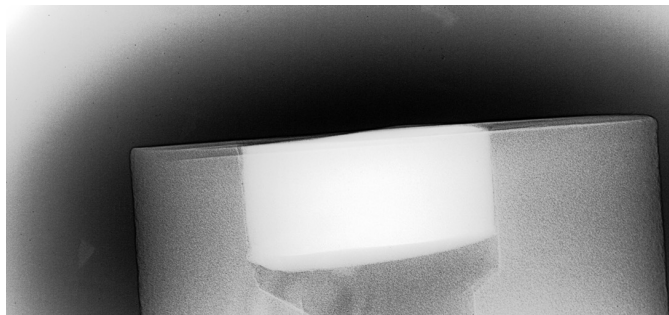
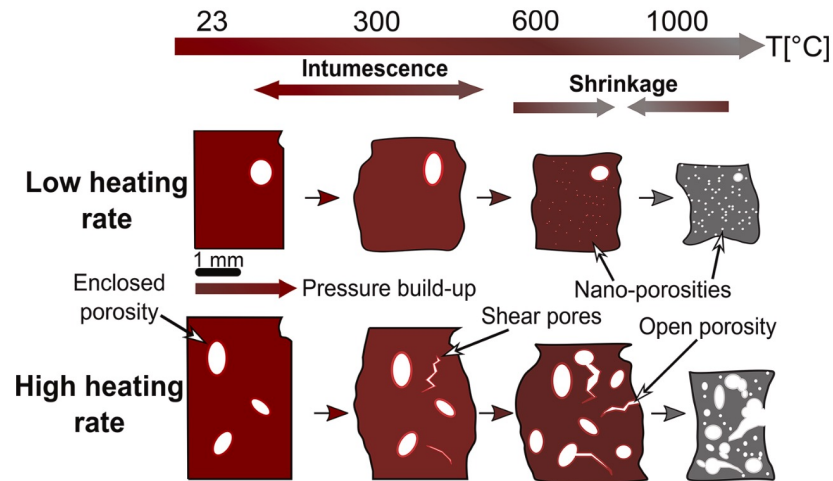


56 °C/min



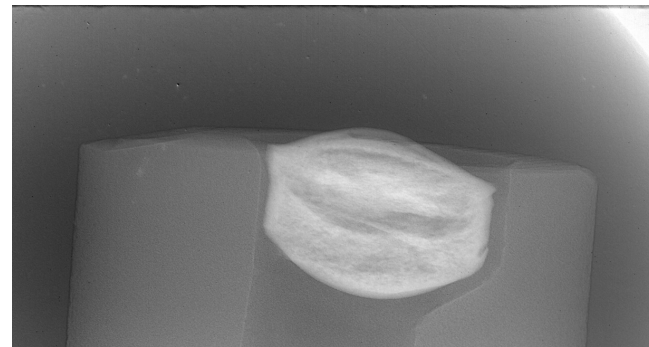
# Exploring higher heating rates with Radiography

- Tomographies indicated heating-rate dependent behavior
- But do not show swelling to the extent seen in plasma wind tunnel tests
  - Heating rates for in-situ tests are significantly lower than for wind tunnel tests/flight.
  - To test at much higher heating rates, radiography was used.

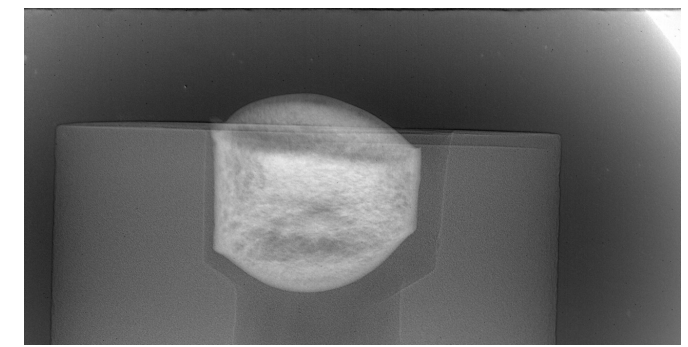


Initial radiograph of RTV sample before high-heating rate tests

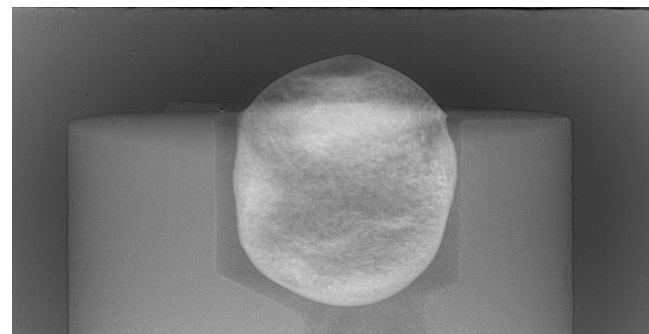
100 C/min



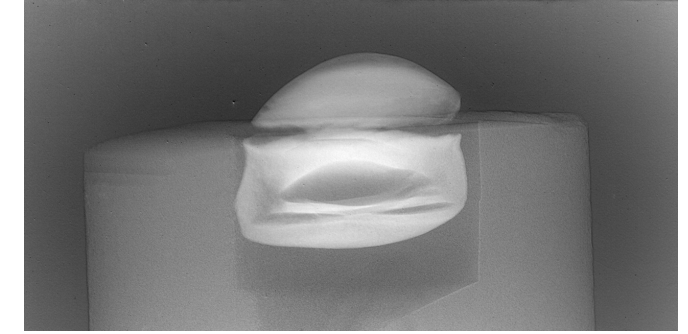
200 C/min



600 C/min



1600 C/min

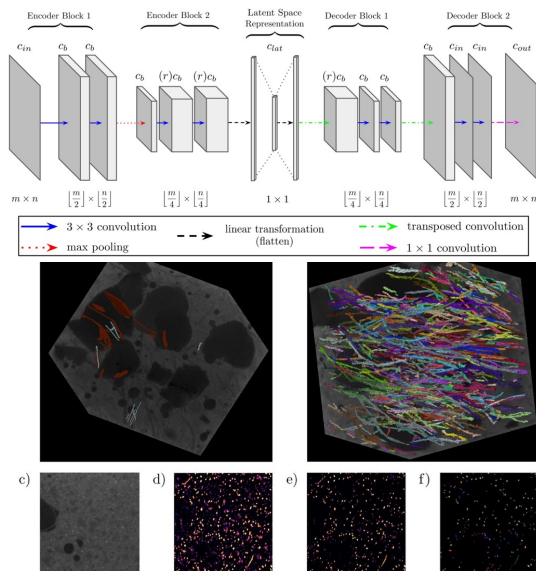


# Conclusion and Future Work

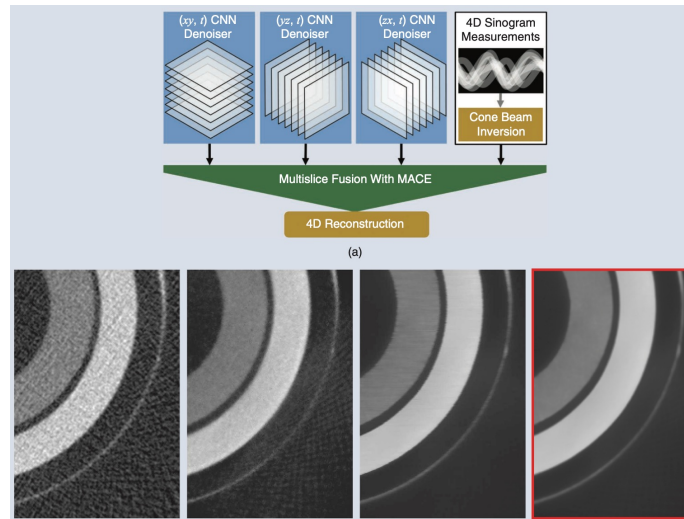
- In-situ pyrolysis of RTV was performed and quantitative determination of morphologic showed:
  - A large change in RTV's porosity and permeability with increase in temperature,
  - Extent of swelling and shrinking is a function of heating rate.

## ➤ Future Work:

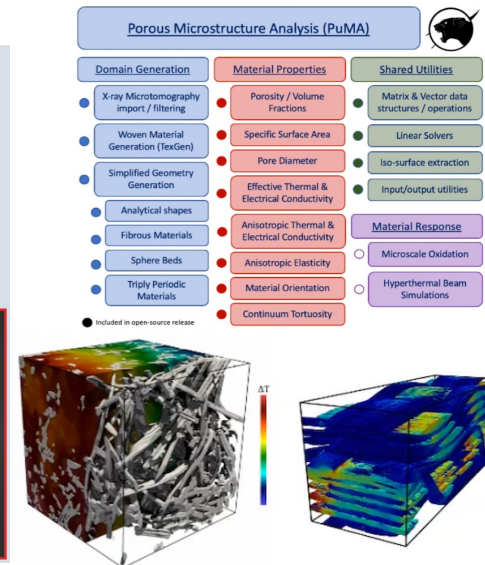
### High-fidelity Segmentation with DLSIA



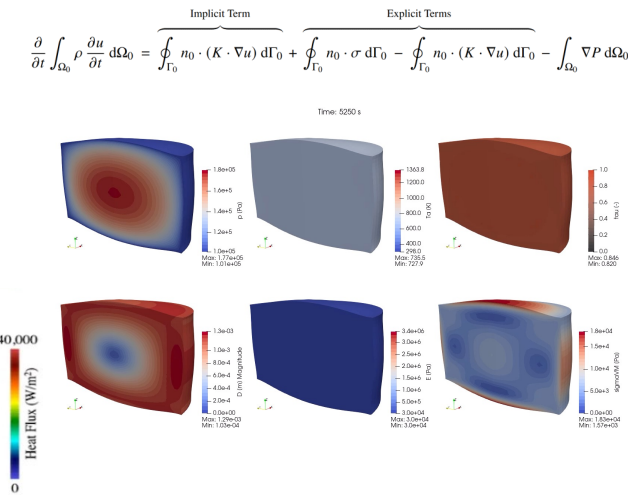
### Increased temporal resolution with MBIR



### Microscale properties with PuMA



### Macroscale simulations with PATO





# Acknowledgements

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# References

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2. Beck, Robin AS, et al. "*Development of the mars science laboratory heatshield thermal protection system*", 2014.
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Thank you! Any questions or suggestions?

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