

AVIATION

8–12 JUNE 2026 | SAN DIEGO, CA





09 JUNE 2026

Extending mARC II Arc-Jet Test Duration via Design and Implementation of Model System Cooling Sleeve

Kate Edwards: Mechanical Engineer, Sierra Lobo Inc.

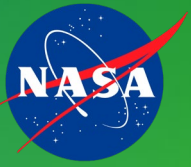
Jocelino Rodrigues: Research Fellow, NASA Ames via ORAU

Daniel Philippidis: Facility Engineer, NASA Ames

NASA Ames Research Center, Thermophysics Facilities Branch



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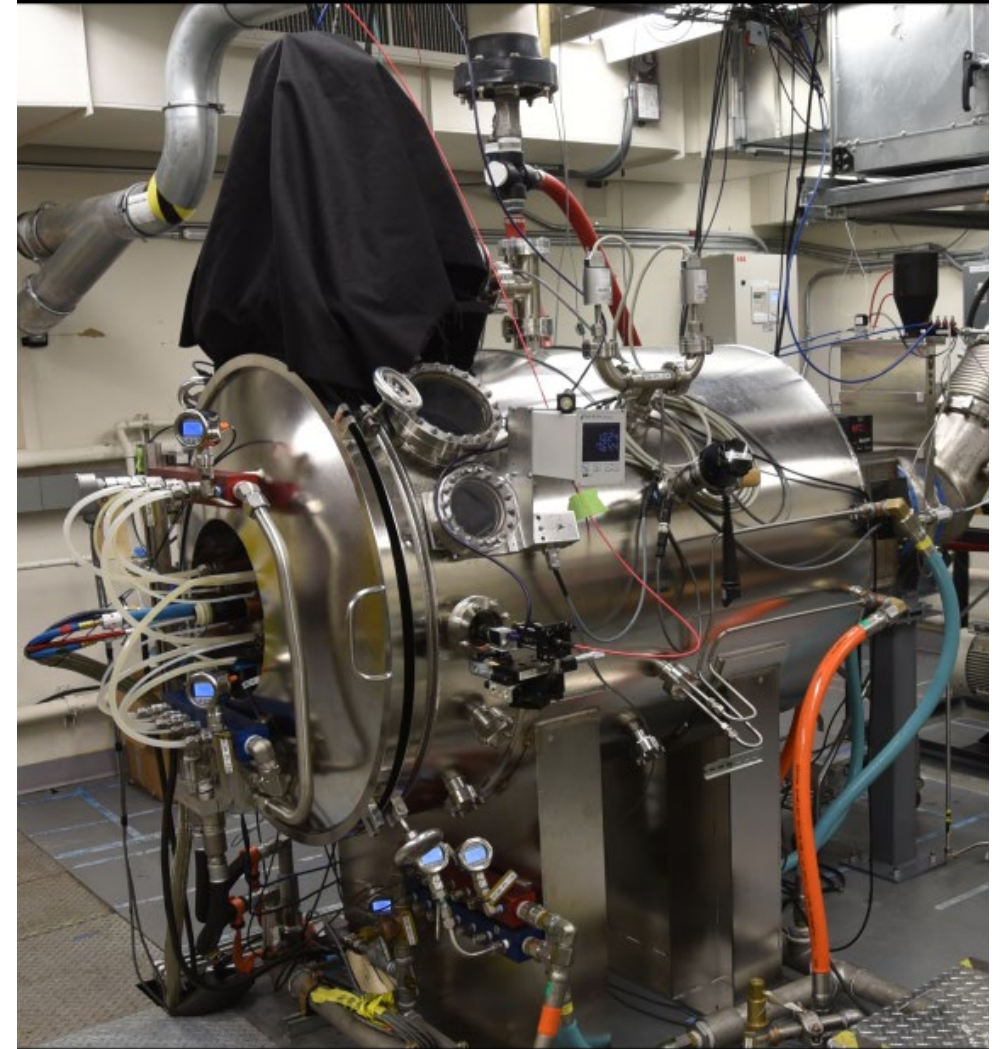


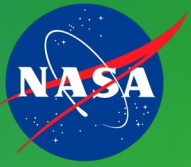
Facility Overview



mARC Overview

- Up to 4-30kW power
- Up to 69mm model position from nozzle
- Up to 1" diameter models
- Two motors for model positioning
- See previous AIAA papers for more details:
 - MacDonald 2023 (AVIATION)
 - MacDonald 2019 (AVIATION)
 - Rodrigues 2024 (AVIATION)
 - Palmer 2025 (AVIATION)
 - Rodrigues 2025 (SciTech)





Project Objectives





Uncooled Model System

- Heat transfers from model, though swing arm, to shaft and feedthrough.
- Ferrotec ferrofluidic feedthrough used for vacuum sealing motor shaft.
- Max operating temperature of 80°C.



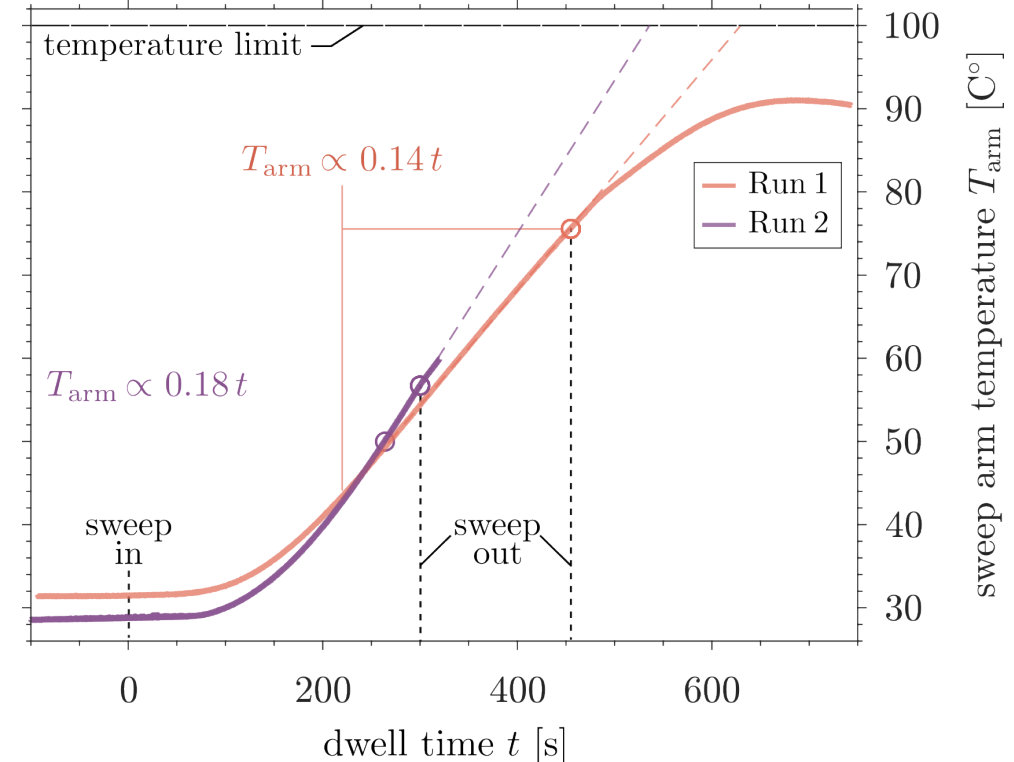
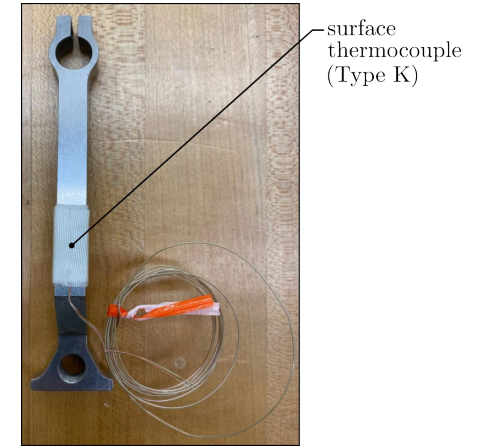
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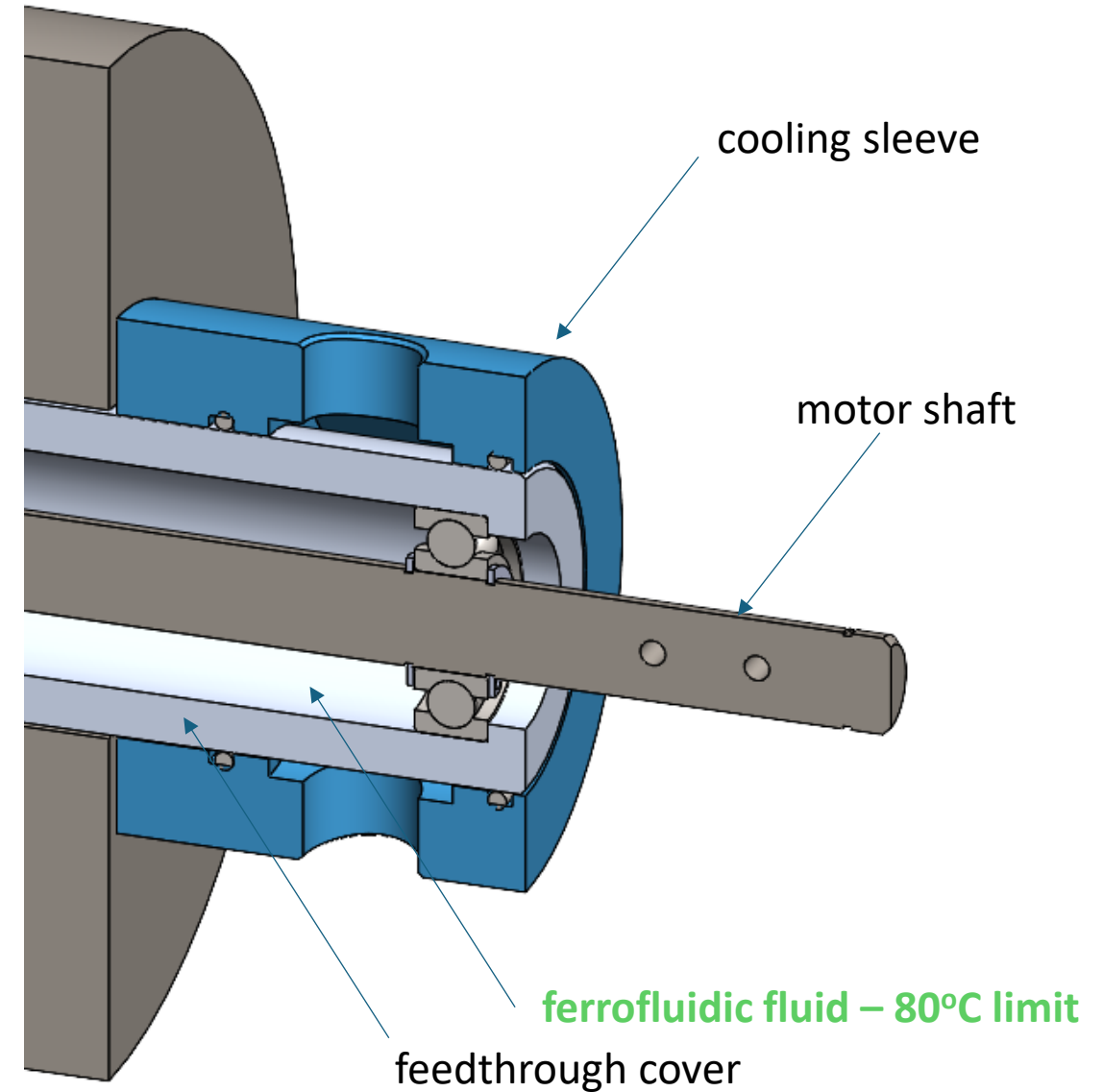
Initial Heating Data

- Thermocouple data at support arm to find max run time.
- Limited to 450s or 7.5m with self-imposed 100°C limit at support arm.
 - Minimum power 4kW (current and flow rate)
 - Maximum distance
- System insulated with Refrasil



Proposed Solution

- Aluminum sleeve to be installed over feedthrough.
- Advantages:
 - Fast turnaround
 - Simple design – no moving parts
 - Inexpensive
 - Balance of conductivity and strength
- Disadvantages:
 - Small cooling area
 - Far from heating source





Analysis Overview



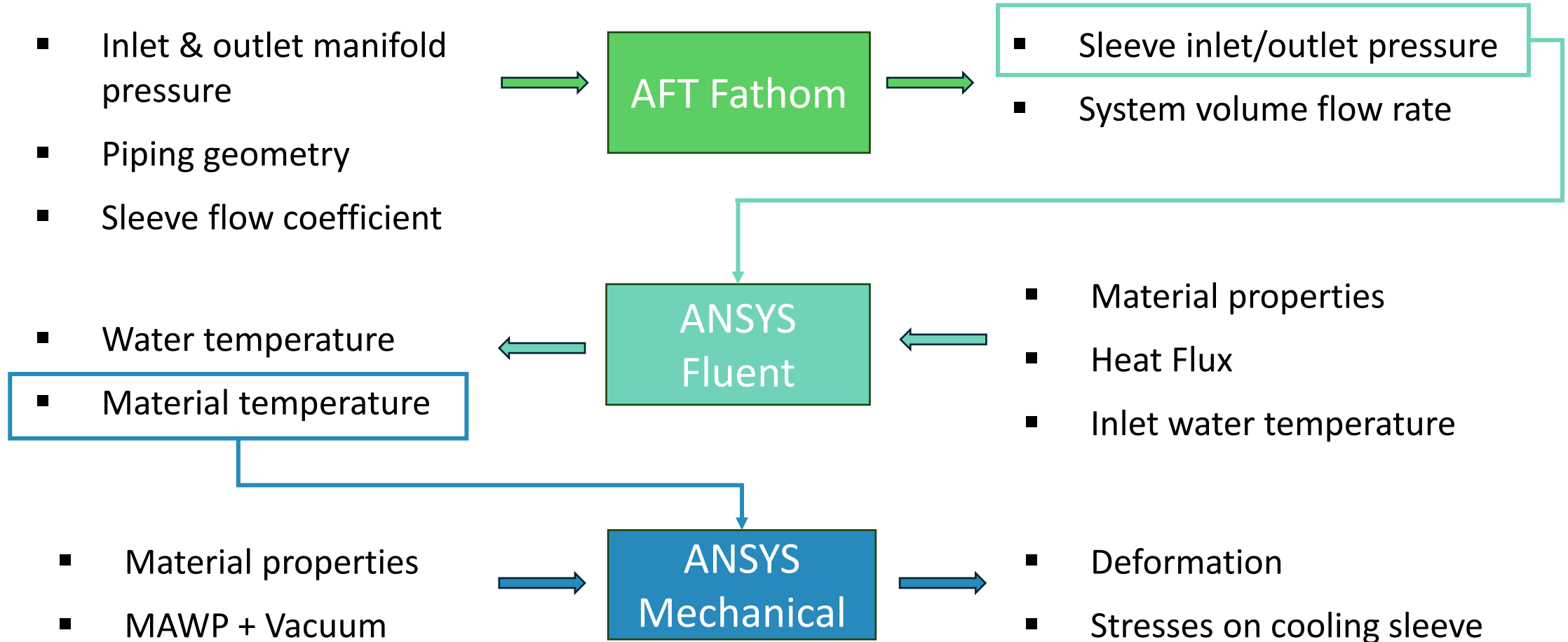


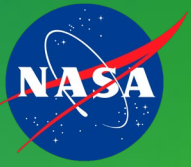
Big Questions

- Will the existing pump be able to handle the additional flow? → ■ AFT Fathom
 - Incompressible Flow Solver
- Will the sleeve sufficiently cool the feedthrough at highest heating conditions? → ■ ANSYS Fluent
 - Steady State Fluid Flow
- Will the sleeve be able to mechanically withstand the pressure/temperature? → ■ ANSYS Mechanical
 - Static Structural FEA



Analysis Map





Piping Analysis

AFT Fathom

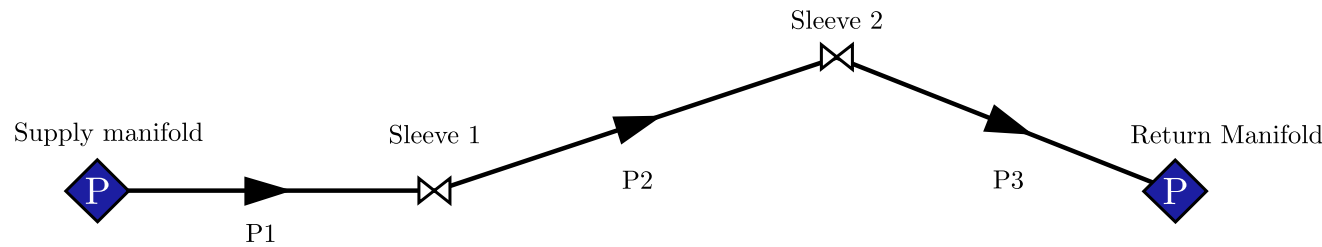


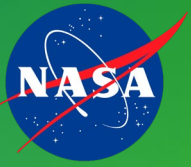
Setup

- Input supply & return pressure
- Piping material, lengths, heights, inner diameters, bends
- Sleeve flow coefficient
- $C_v = K\dot{V} \sqrt{\frac{SG_w}{\Delta P}}$

Results

- Flow rate
- Sleeve inlet & outlet pressure
- $\Delta P_{Sleeve1} = \Delta P_{Sleeve2}$
- $error_{C_v} = \frac{C_v^* - C_v}{C_v} = .16\%$





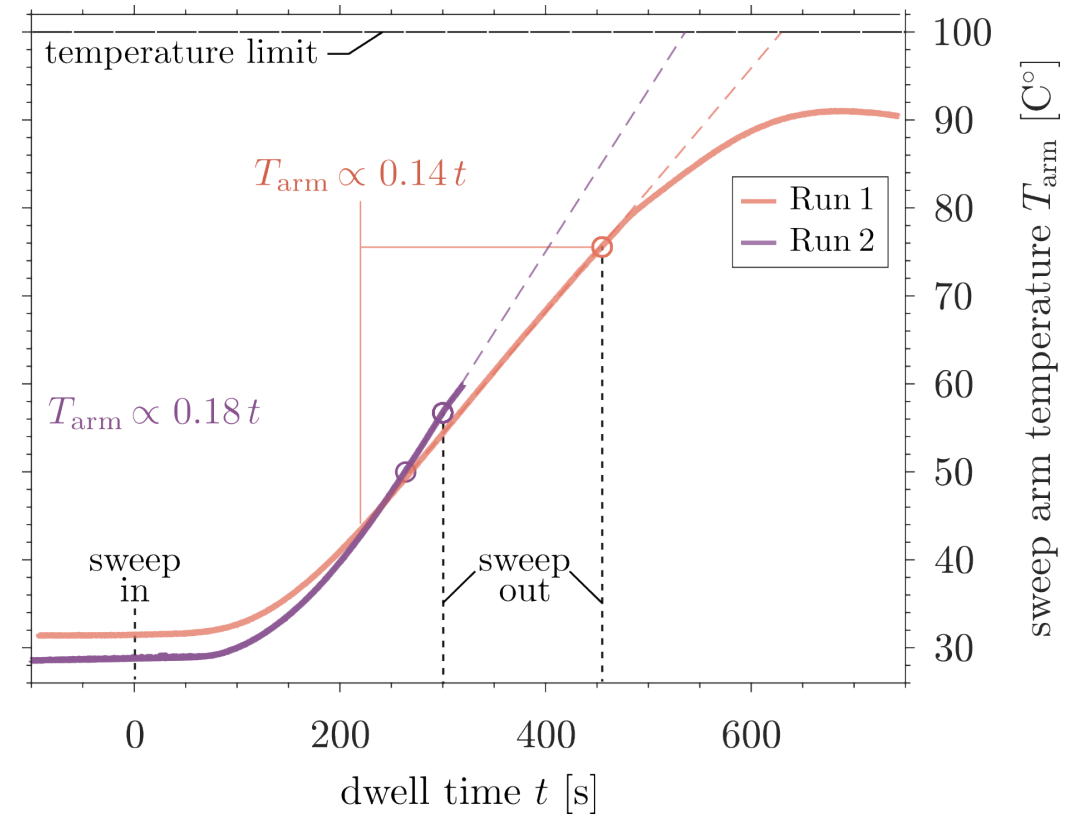
CFD & Thermal Analysis

ANSYS Fluent



Inputs

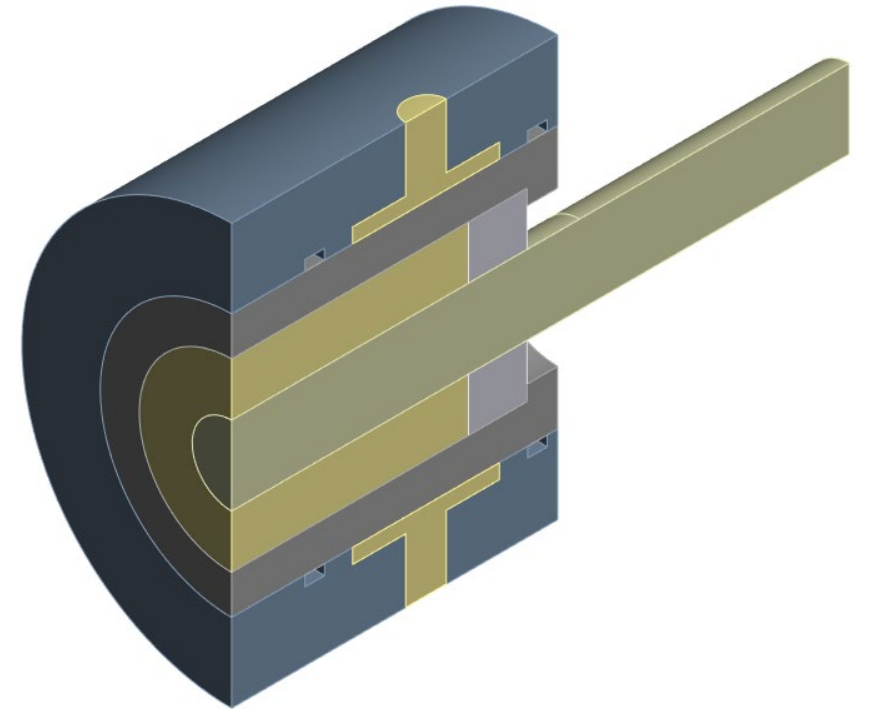
- Water inlet temperature
- Material properties
- Heat flux
 - $Q = mc_p\Delta T$
 - $\dot{Q} = \frac{Q}{350s}$
 - $\dot{Q}_{max} = \dot{Q} * \frac{30kW}{4kW}$
 - $\dot{q} = \frac{\dot{Q}}{A} = 45 \text{ kW/m}^2$





Model Assumptions

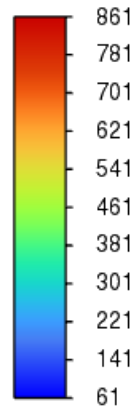
- Only model section within test box
- Assume bearing as solid steel
- Shaft is only heated surface



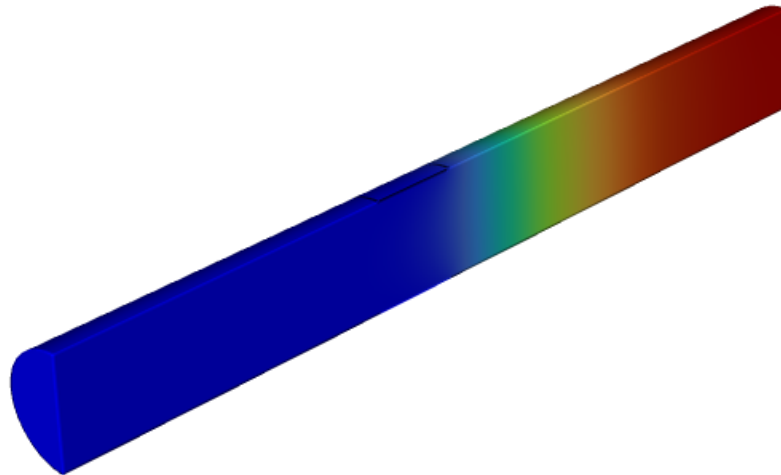
Results

- Ferrofluid all remains below 80°C
- Edge of bearing reaches 120°C
- End of shaft reaches 860°C

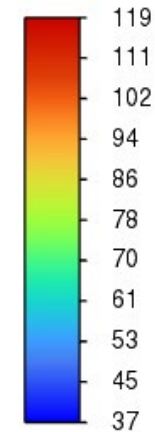
Static Temperature
[C]



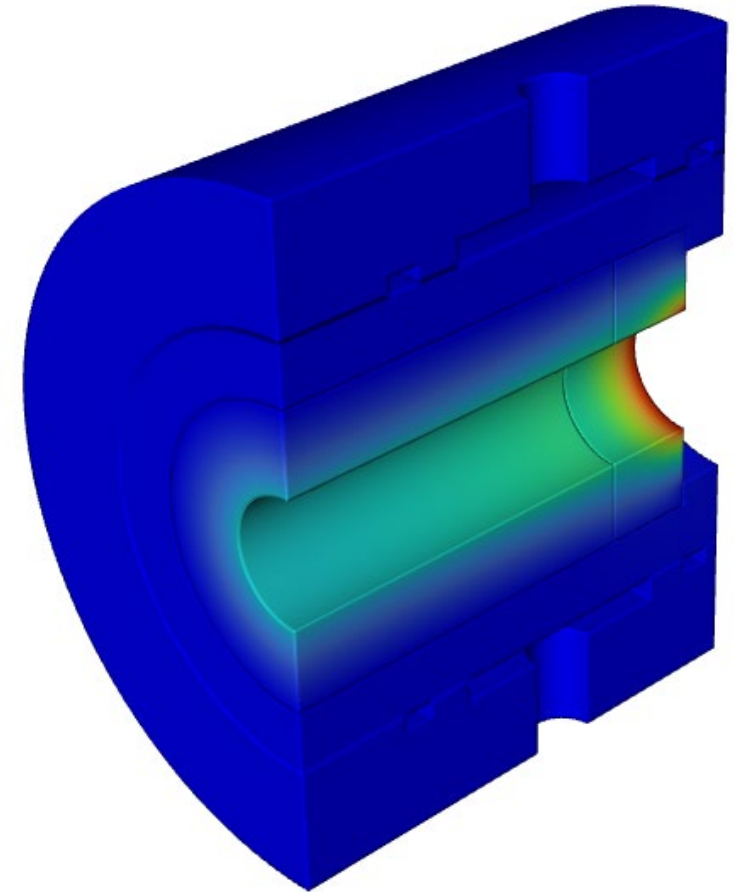
shaft-temp

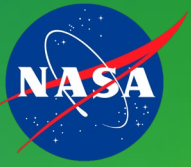


Static Temperature
[C]



bearing-temp





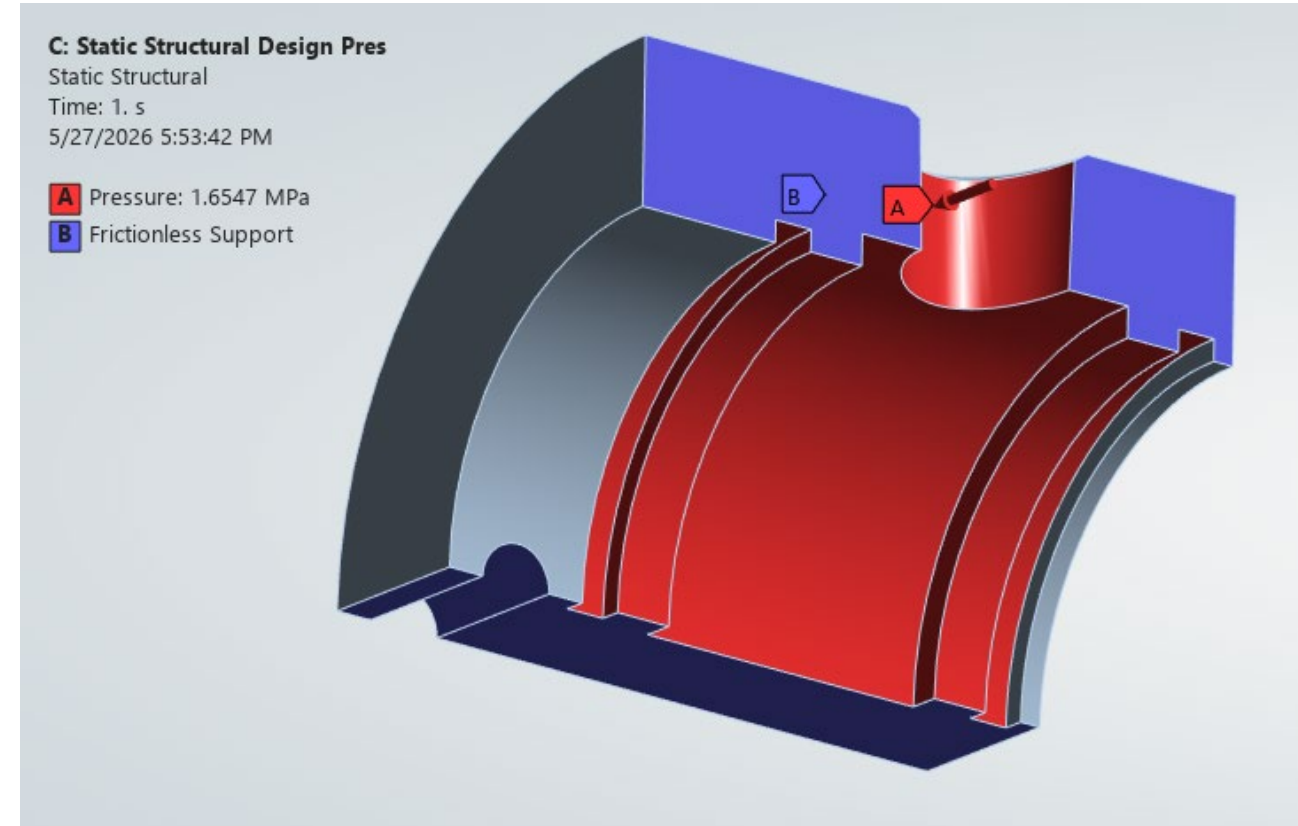
Static Structural FEA

ANSYS Mechanical

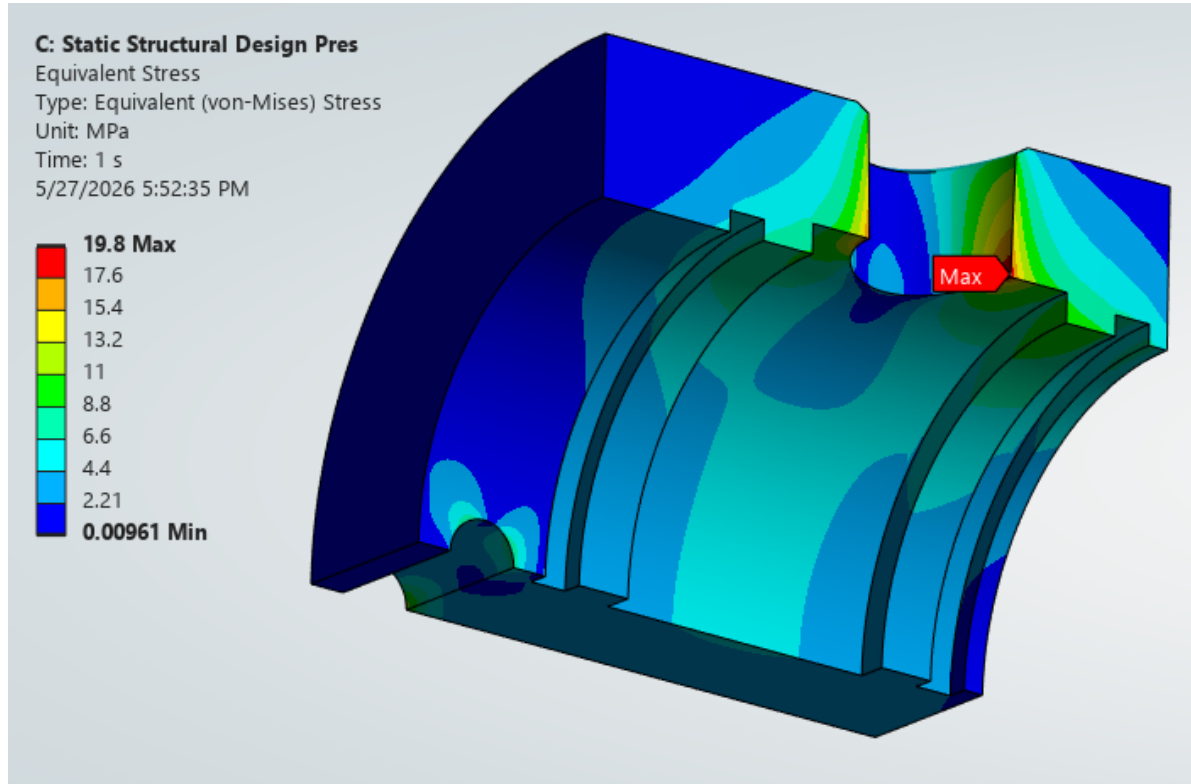


Boundary Conditions

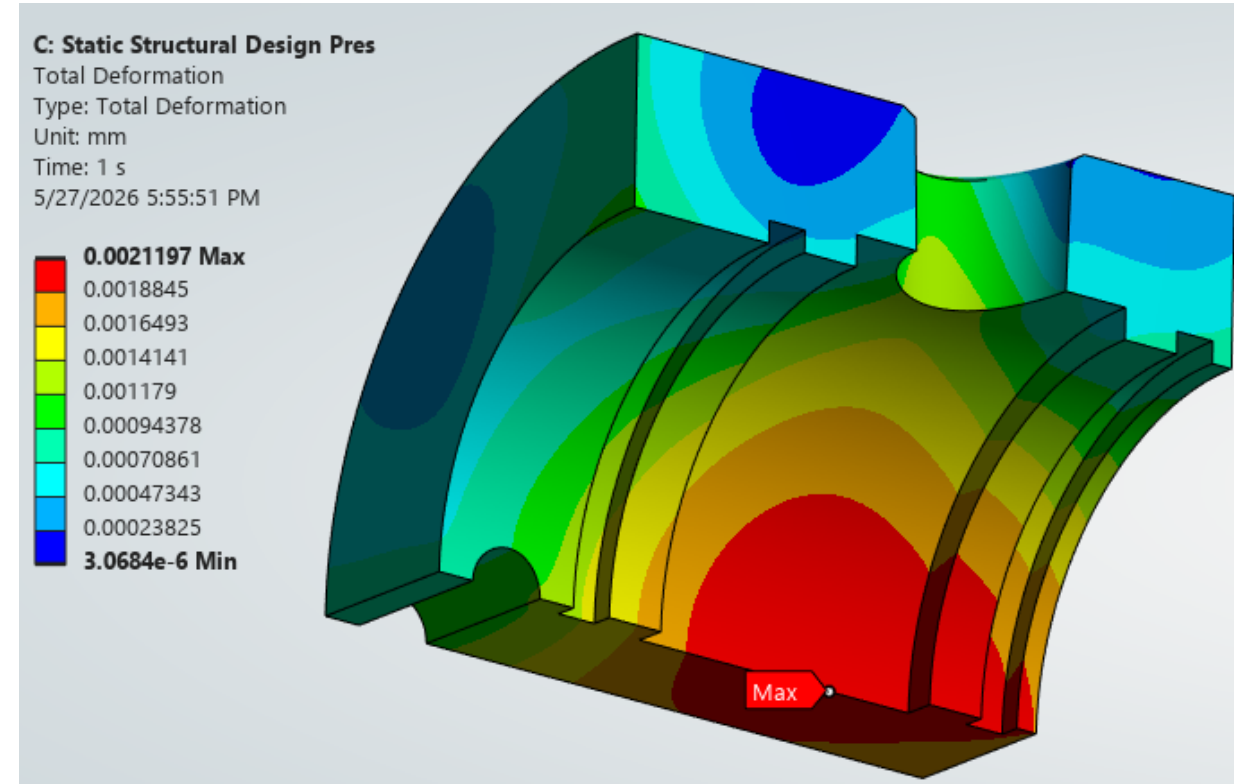
- Frictionless support
 - Quarter symmetry
 - Set screw
- MAWP + Vacuum
- Elastic modulus at maximum temperature used
 - Stays near room temperature



Results



19.8 MPa less than ASME allowable 75 MPa



2.2×10^{-3} mm deformation



Installation and Outcome



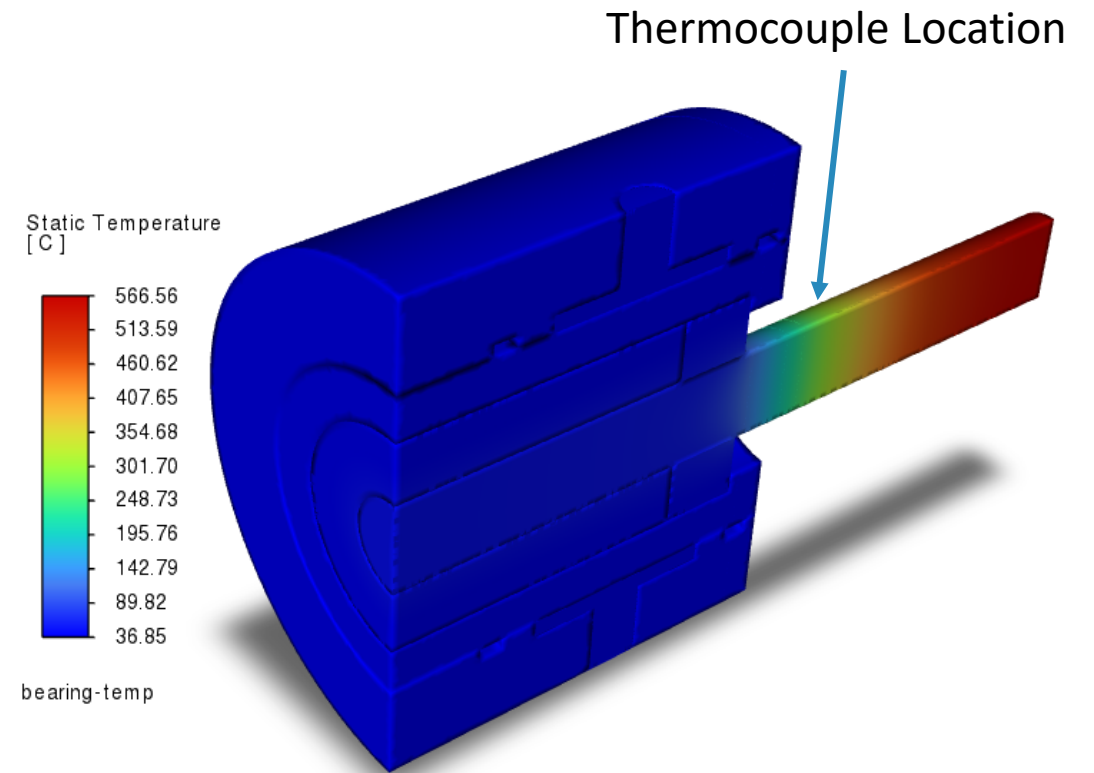
Two Sleeves Successfully Installed

- Pressure test passed
- Ran Space X and Blue Origin tests successfully
- Durations up to 750s at higher power and closer to the nozzle
- Placed thermocouple between feedthrough and model support



Further Analysis

- Thermocouple reached steady state at 160°C
- Re-ran CFD analysis to reflect test conditions
- Analysis thermocouple temperature was higher
 - 247°C
- Points to conservative analysis as expected





Future Work





Operational

- Install flow meter
 - Operational need to guarantee cooling
- Extra surface thermocouple
 - Inform future modeling and understanding of heating

Design

- Short term
 - Increase number of models held by support arm
- Long term
 - Remote axial movement
 - Expanded cooling

Come Test with Us at the Mini Arc!

To schedule testing at the arc jets:

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- Email: megan.e.macdonald@nasa.gov

For general inquiries of our large arc jets:

- **Frank Hui**
- Phone: 650-604-5395
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Peter Race

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TSF Website:

<https://www.nasa.gov/thermophysics-facilities-branch/>

