



TFAWS
MSFC • 2026

Predicting Operational Performance of xEMU Boot at Lunar South Pole Temperatures using Thermal Desktop®

Emma Quick, *Amentum/JSC*

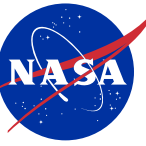
Noah Andersen, *HX5/JSC*

Presented By
Emma Quick

Thermal & Fluids Analysis Workshop
TFAWS 2026
August 31 – September 4, 2026
NASA Marshall Space Flight Center
Huntsville, AL



Agenda



- Background and Objectives
- Test Setup/Conditions/Instrumentation
- MiniShroud Model Overview
- MiniShroud Model Correlation
 - Results, performance
- Boot Model Overview
- Boot Model Correlation
 - Results, performance, RMSE
- Lunar Operation
 - Modeling, heat flux and temperature predictions
- Conclusions

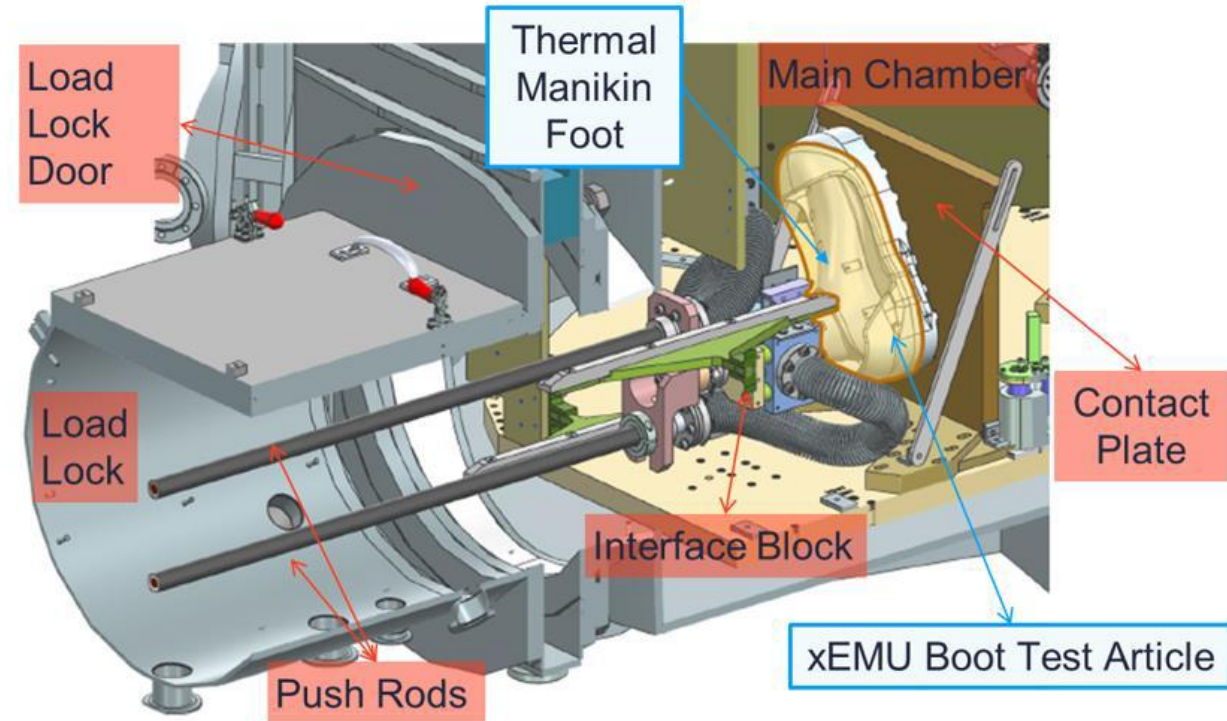
Background & Objectives

- Upcoming Artemis missions will land astronauts on the moon at the south pole
 - Temperatures here can be as low as ~ 50 K
- Exploration Extravehicular Mobility Unit (xEMU) boots will need to withstand these temperatures throughout extravehicular activity (EVA)
 - Comprehensive TVAC testing and thermal model correlation of the xEMU boot are essential to ensure crew safety
- Test data was used to correlate the thermal model and subsequently predict lunar boot performance in permanently shadowed regions on the moon



Test Setup, Instrumentation, and Conditions

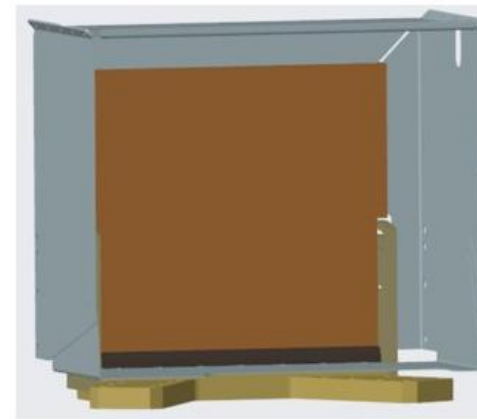
- CITADEL TVAC chamber at NASA JPL
- Thermal manikin foot outfitted with xEMU boot
 - Water lines running through the foot maintain a skin boundary temperature
 - Thermocouples/RTDs on various layers of the foot+boot stack-up
 - 56 external and 16 internal TCs
 - Ventilation flow of 12.4 SLPM (1.5 ACFM) between the sock and bladder layers
 - Pressurization of boot to 4.3 psia
- “MiniShroud” for environmental temperature control
 - Aluminum stomp pad for boot to contact
 - Six environment TC coupons
- Boot mounted on actuated push-rod system to simulate IVA to EVA transition



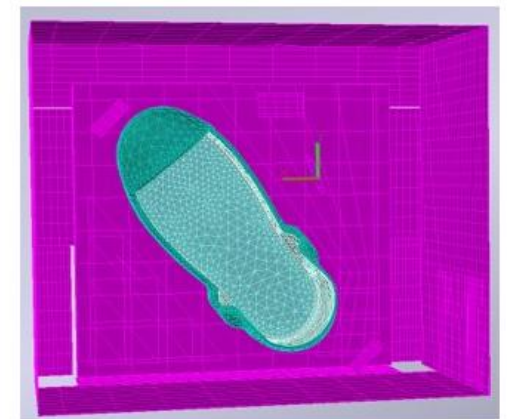
Test Parameters:

Variable Name	Levels	Units
Contact Pressure	0, 40, 100	lbf
Ventilation Flow	0, 1.5	ACFM
Foot Temperature	28, 33	°C
Environment Temp	48, 100	K

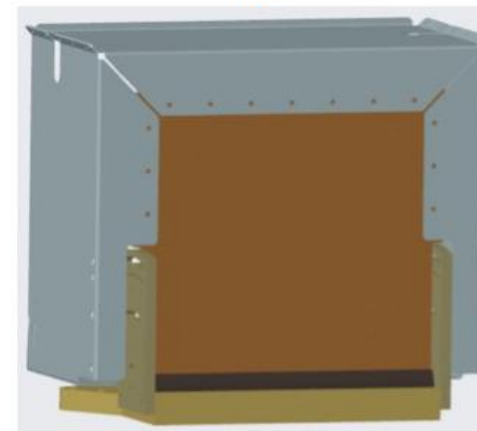
- It was necessary to build the MiniShroud model instead of using single external sink node for the boot
 - In test, the MiniShroud did not provide a cold, isothermal environment
- Model built from JPL CAD and includes stomp pad, top/side/bottom panels, copper bars, Teflon stomp pad coverings, and SLI curtain
- External MiniShroud environment simulated with sink temperature nodes on each face
- G10 spacers on bottom of MiniShroud were not explicitly modeled, but represented with a gap and low-conductance contactor
- Mechanical joints (screws, panel overlaps, interfaces) were modeled with face/edge contactors
 - Conduction coefficients calculated based on screw count, conductivity, and contact area



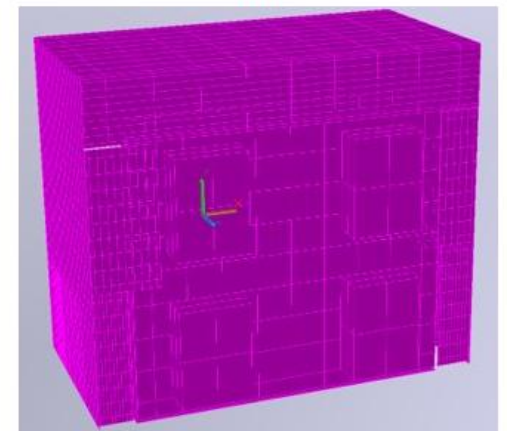
(a)



(b)



(c)



(d)

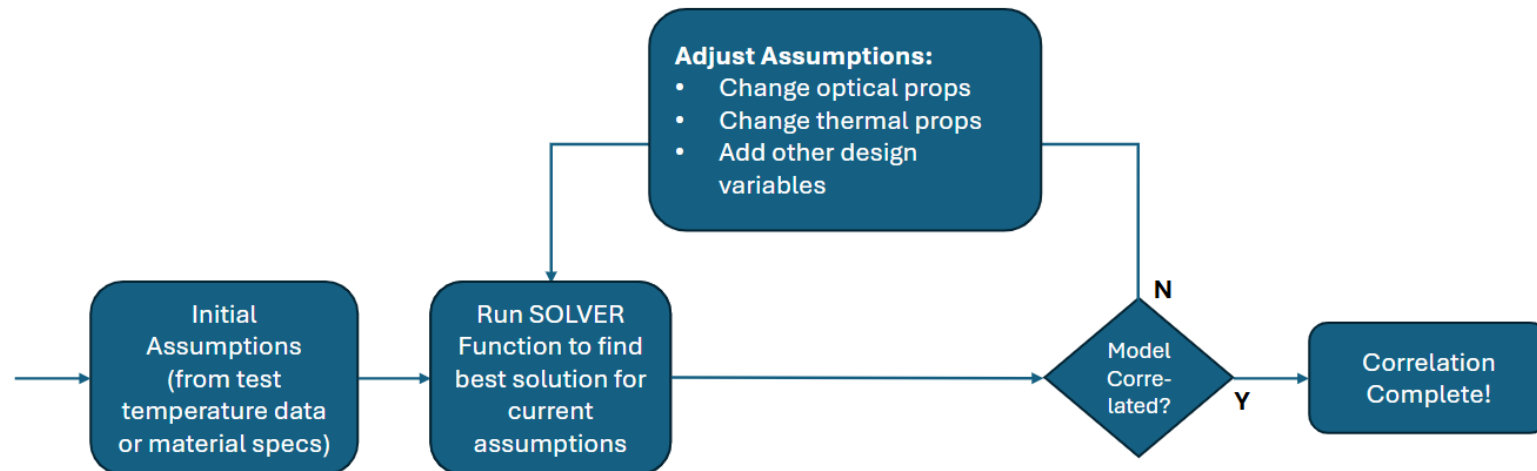
Comparison of CAD file (.stp) and Thermal Desktop (TD) geometry. (a) CAD interior, (b) TD interior, (c) back of CAD, (d) back of TD.

- The variables with highest uncertainty in the MiniShroud model were fed into Thermal Desktop SOLVER as design variables
 - External environment sink temperatures, conductance from stomp pad to MiniShroud floor, conductance from bottom panels to the sides
- Data from the test or from material specifications used as initial guesses in the model
- SOLVER iterates until objective function is minimized. The object is defined as follows:

$$OBJECT = \sqrt{\sum (T_{model,i} - T_{test,i})^2}$$

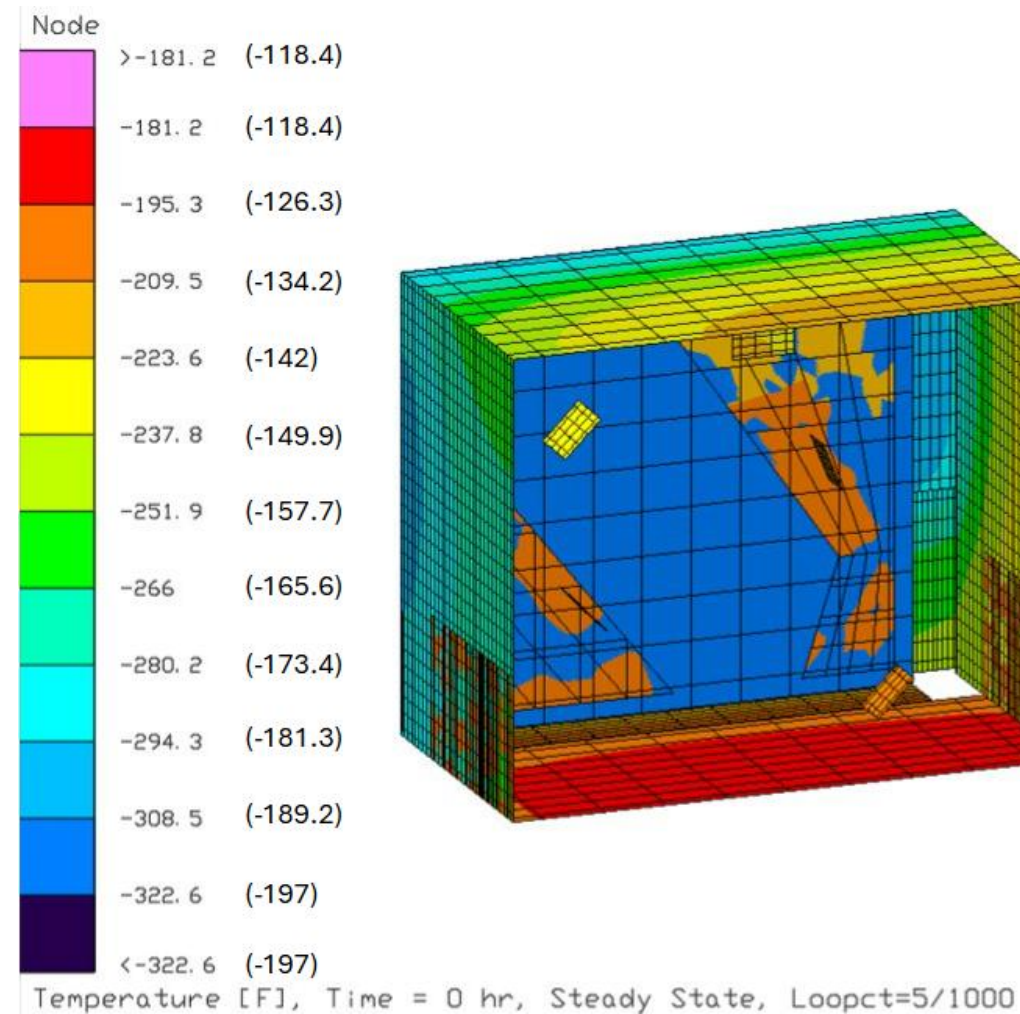
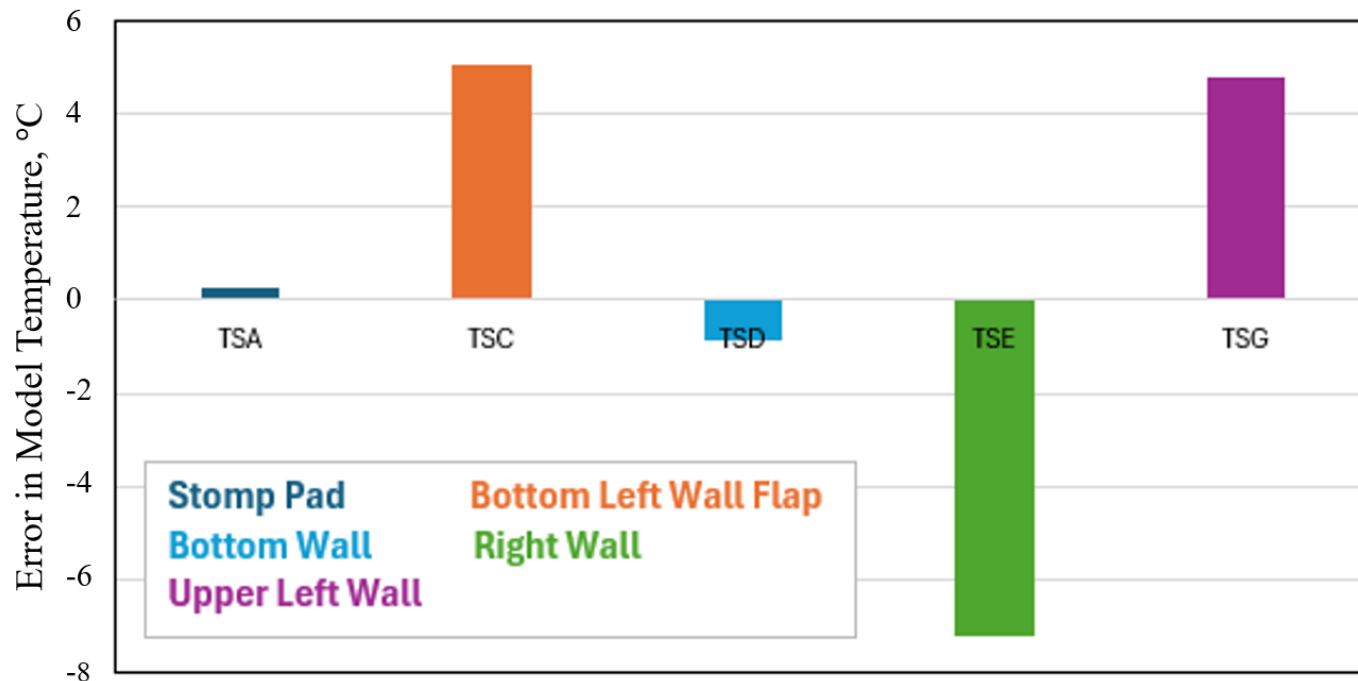
Where T represents temperatures at known TC locations on the walls of the MiniShroud

- Optical and thermal properties were adjusted manually between iterations
- Model was correlated to 4 steady state radiation and contact test points separately since the temperature of the stomp pad affects the wall temperatures

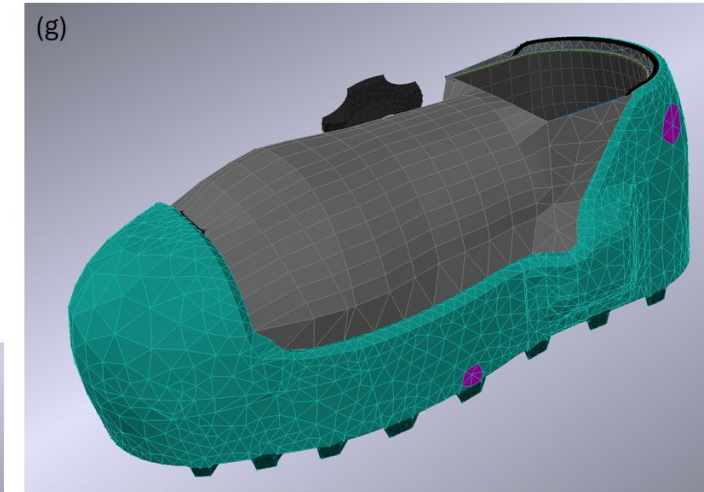
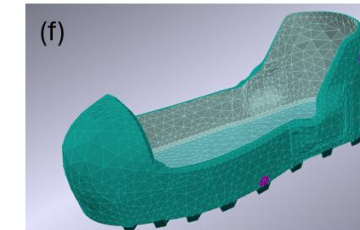
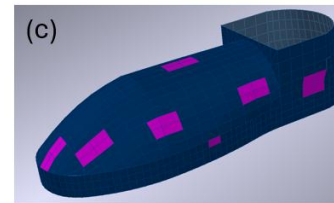
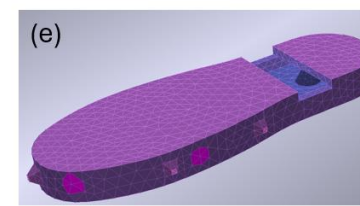
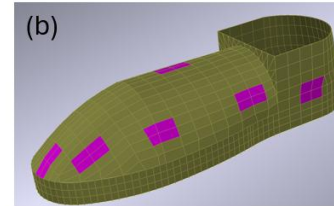
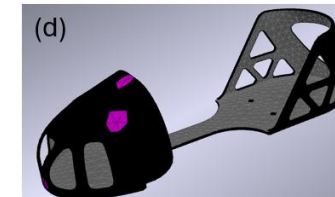
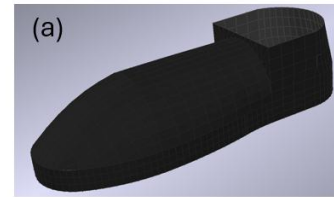


MiniShroud Model Correlation Results

Design Variable	All Radiation Test Points	Contact, 48-60 K setpoint	Contact, 100 K setpoint	Units
Tchambersink_floor	255.8	272.0	264.2	K
Tchambersink_left	284.9	279.8	282.9	K
Tchambersink_right	255.4	255.4	255.4	K
Tchambersink_top	309.1	289.2	299.2	K
Tload	292.0	294.8	294.7	K
StompPad2Bottom	16.139	21.276	25.297	W/K/m ²
Flap2SideScrew	0.100	0.129	0.100	W/K



- Built from CAD of xEMU boot model
- Layer material properties such as conductivity, specific heat, and optical characteristics match actual hardware
- Foot is represented with boundary nodes to match manikin's isothermal behavior
 - Full human foot model not available
- Linear conductors and contacts capture internal heat transfer paths
- External heat transfer paths include radiation and outsole conduction
- EPG regions had adjustable effective emissivity for tuning purposes



Thermal Desktop Boot Model: (a) sock, (b) pressure bladder, (c) restraint layer, (d) frame, (e) thermal insert, (f) outsole, and (g) full assembly. Magenta regions represent TC locations.

- Stomp contact could be modeled with just the tread tops (hard flat surface) or the entire outsole (granular regolith)

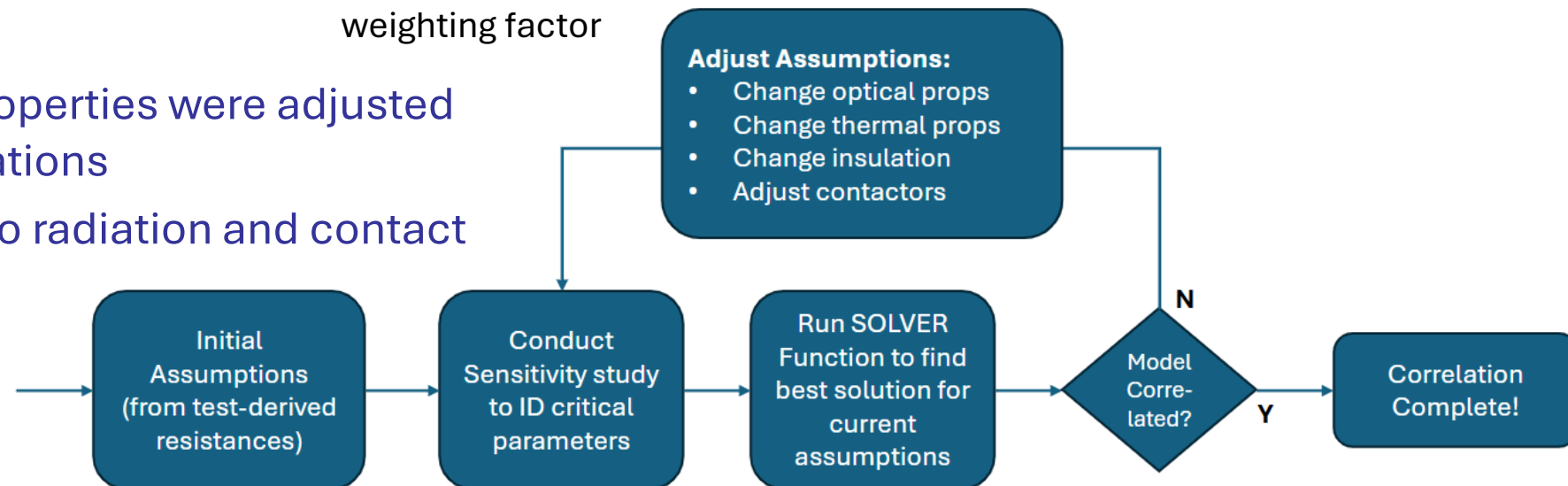
Boot Model Correlation

- The variables with highest uncertainty in the boot model were fed into Thermal Desktop SOLVER as design variables
 - Thermal contactors/conductors between boot layers, thermal and optical properties, and insulation performance
- Data from the test (calculation conductances) used as initial guesses in the model
- SOLVER iterates until objective function is minimized. The object is defined as follows:

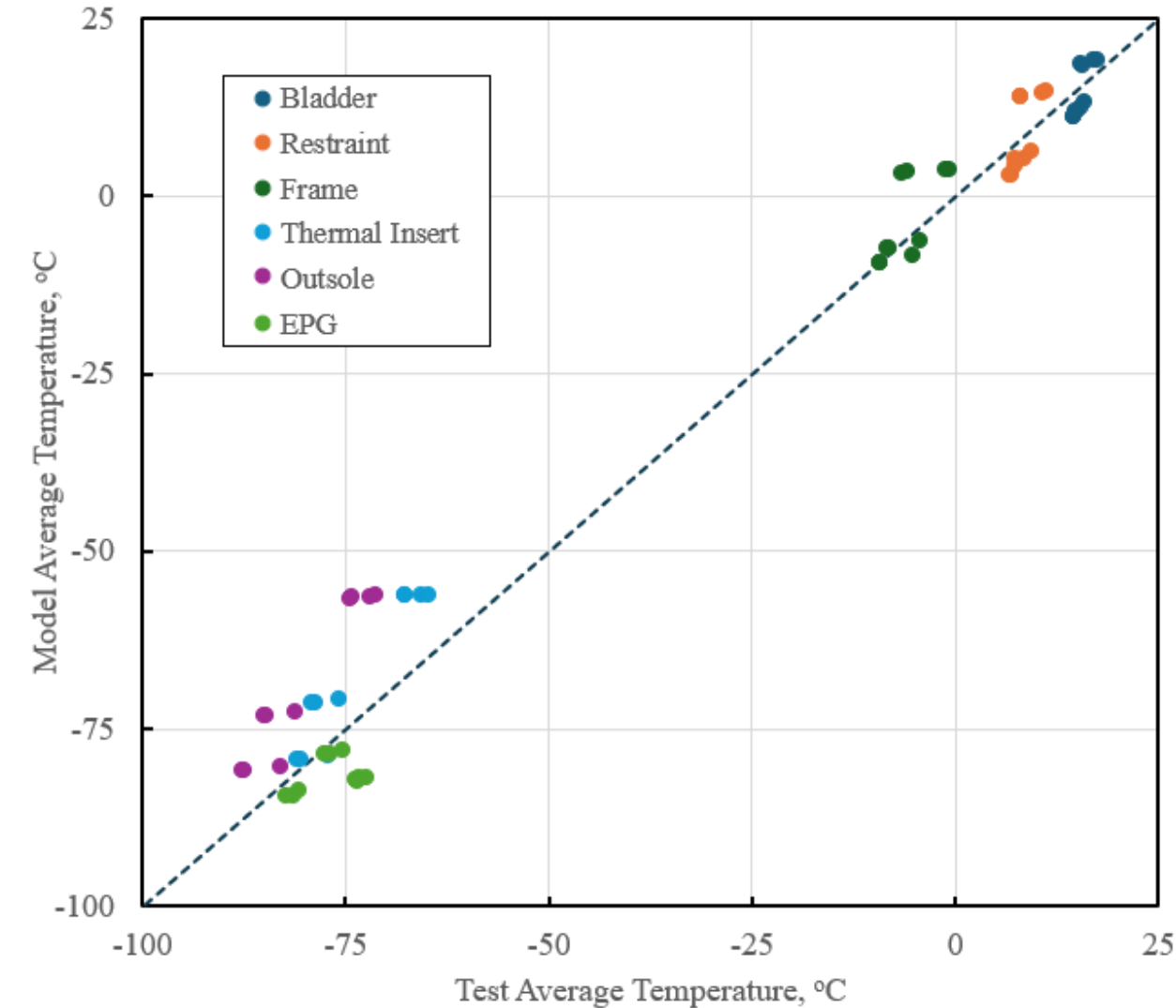
$$OBJECT = \sqrt{\sum (T_{model,i} - T_{test,i})^2 + K \cdot \sum (q_{model,i} - q_{test,i})^2}$$

Where T represents temperatures at known TC locations on the boot, q represents the heat leak out of the book, and K is a weighting factor

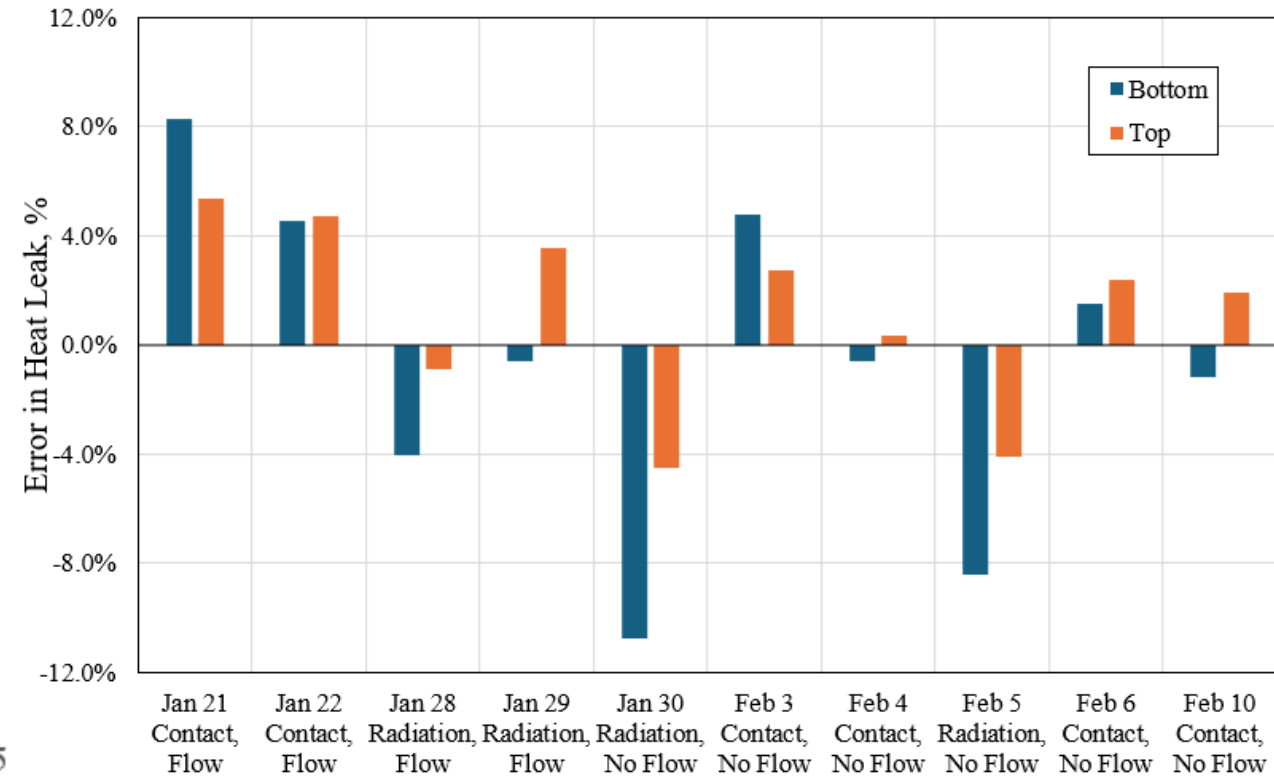
- Optical and thermal properties were adjusted manually between iterations
- Model was correlated to radiation and contact test points separately



Boot Model Correlation Results

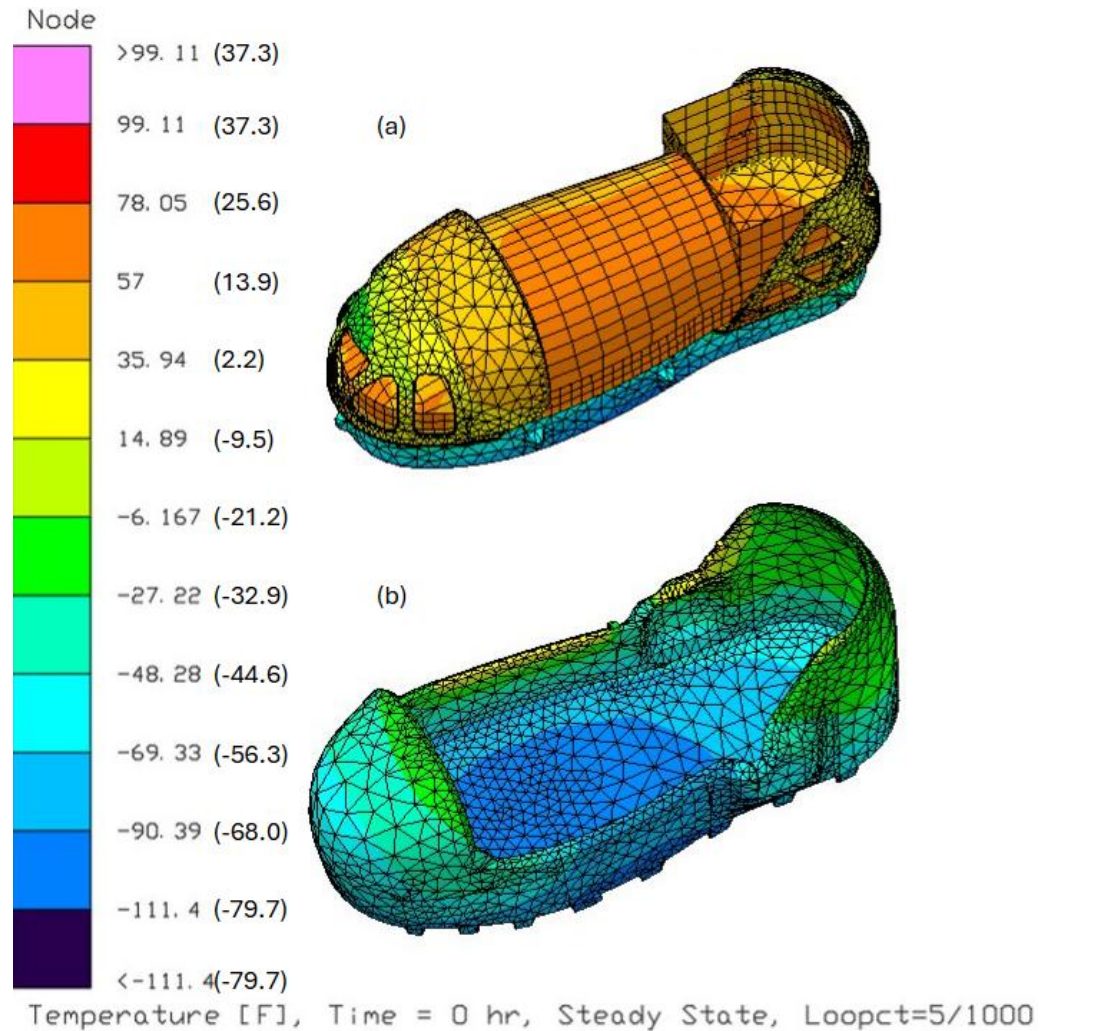


In general, the model is slightly conservative for contact cases (overpredicting heat leak) and slightly non-conservative for radiation-only test points.

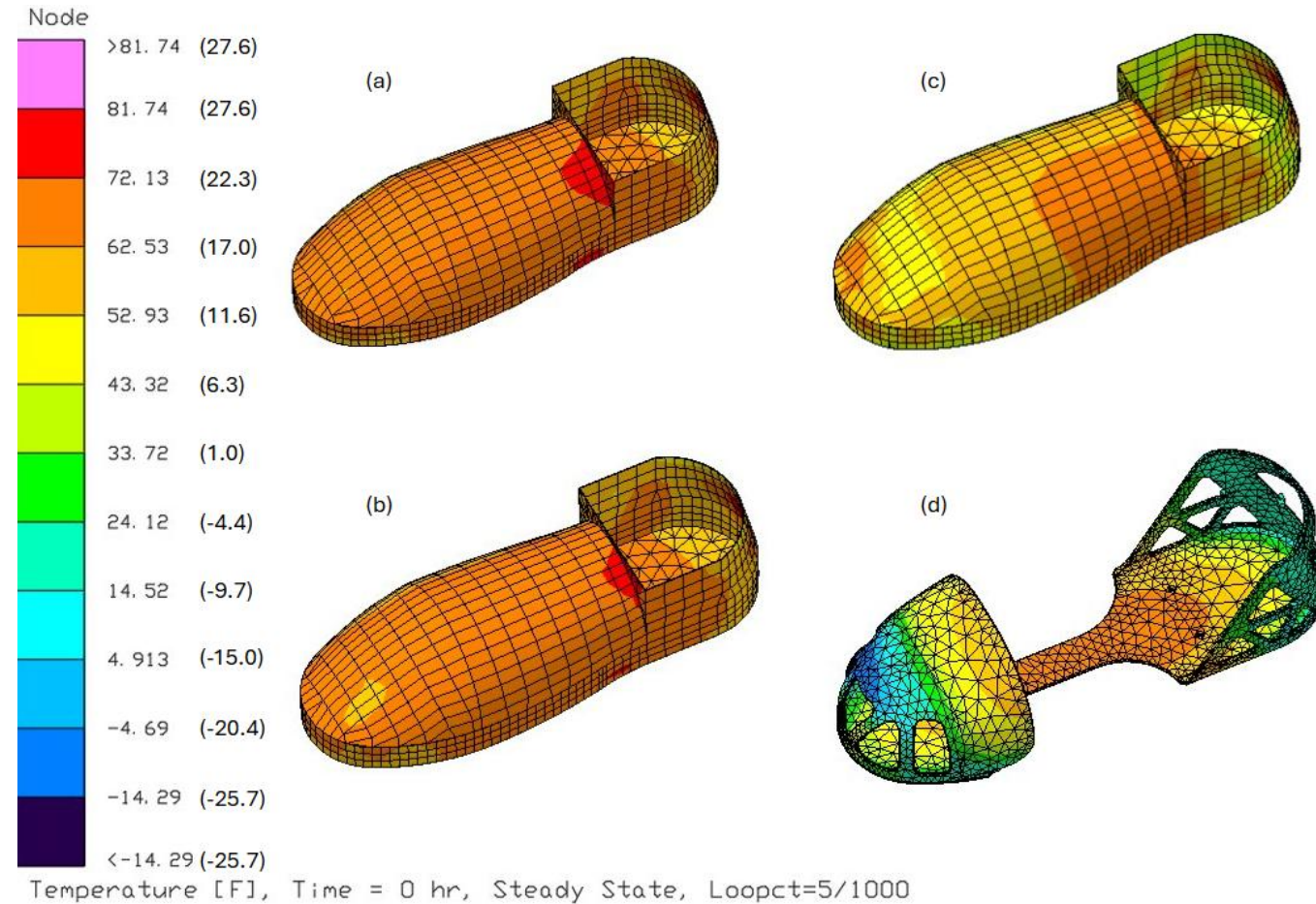


Above line means model is warmer; inner layers are more well correlated

Boot Model Thermal Performance



Radiation case temperature gradients between layers of the boot. (a) restraint, frame, and thermal insert; (b) outsole. Temperatures in Celsius are provided in parentheses.

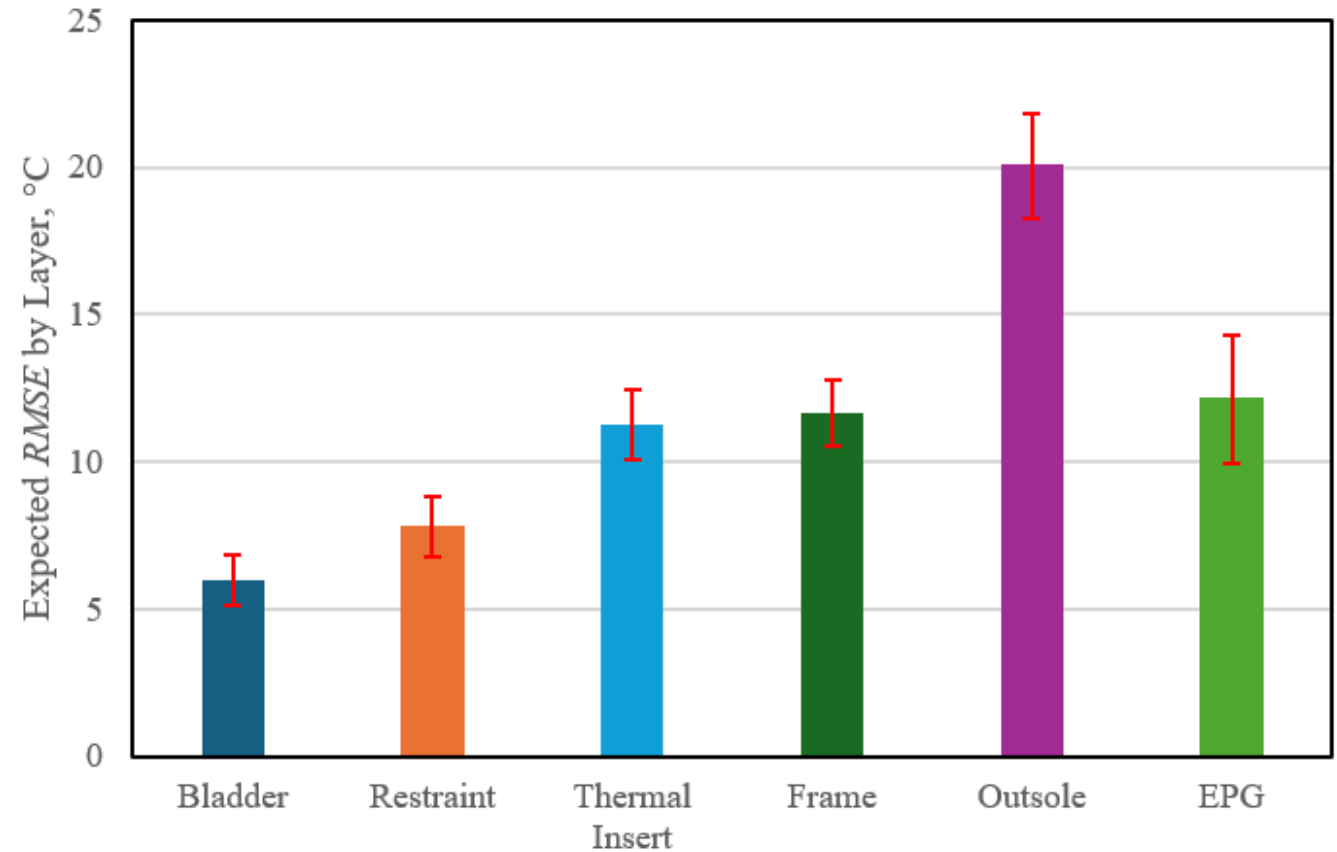


Radiation case temperature gradients throughout individual layers of the boot. (a) sock, (b) pressure bladder, (c) restraint, (d) frame. Temperature in Celsius are provided in parentheses.

$$RMSE_i = \sqrt{\frac{\sum_{i=1}^N (T_{model,i} - T_{test,i})^2}{N}}$$

Sensors closest to the foot exhibit the lowest error, while sensors near the outsole and EPG have higher variability due to large thermal gradients and positional uncertainty.

Heat leak predictions were highly accurate (low RMSE)



Location	Expected RMSE (W)	Uncertainty on RMSE (W)
Bottom of Foot	0.101	+0.042 / -0.033
Top of Foot	0.076	+0.021 / -0.018

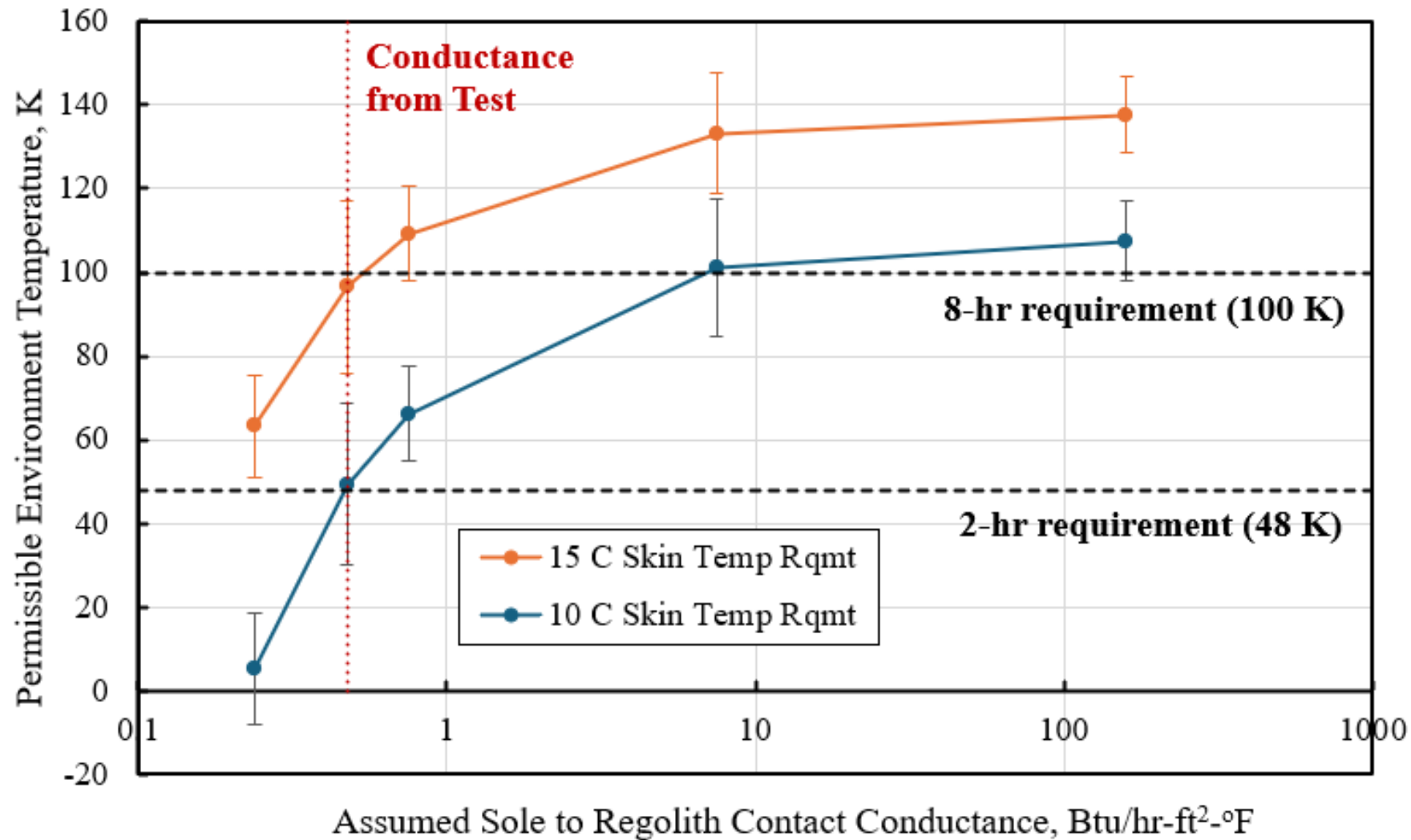


Lunar Operation Modeling



- Gaps identified for accurate representation of lunar operation
 - Unknown outsole-to-regolith thermal conductance
 - Lack of high-fidelity metabolic foot model
 - No dynamic gait modeling such as flexion and compression
- Updates to model:
 - Outsole contact increased to include tread valleys that would be filled by granular regolith
 - Initial conductance came from test-derived outsole-to-stomp pad value of $0.476 \text{ Btu/hr/ft}^2/^{\circ}\text{F}$
 - Foot represented with the upper and lower nodes as well as a new third node representing the toes, since this area was colder throughout testing
- Permissible Heat Flux Criteria:
 - Used to assess safety since model does not resolve detailed foot anatomy
 - Limits derived from EMU glove testing which used high-fidelity metabolic hand model
 - Thumb tip used as analog for toe heat flux requirement
 - Skin temperature thresholds were identified and the heat flux leaving the test article that resulted in the skin temperature hitting that threshold was calculated
 - 10°C (50°F) threshold $\rightarrow 120.8 \text{ W/m}^2$ heat flux
 - 15°C (59°F) threshold $\rightarrow 111.4 \text{ W/m}^2$ heat flux

Lunar Operation Heat Flux Predictions

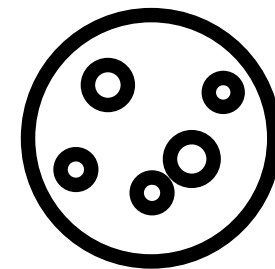


Error bars represent the standard error for the model, as computer using a Monte Carlo Simulation.

- Assumes a constant foot temperature of 10 °C
- Addresses outsole-to-regolith contact conductance uncertainty and inherent model uncertainty

Layer	Minimum Permissible Temperature	Minimum Expected Temperature at 48 K (with 95% confidence)	Minimum Expected Temperature at 100 K (with 95% confidence)
Bladder	-34.4 °C	-37.2 to -5.0 °C	-34.0 to -3.3 °C
Upper Frame	N/A	-109.0 to -37.0 °C	-96.6 to -30.2 °C
Outsole	N/A	-225.2 to -157.7 °C	-173.2 to -126.4 °C

- Derived from contact conductance range of 0.24 Btu/hr/ft²/°F to 159.2 Btu/hr/ft²/°F
- Model uncertainty incorporated by adding RMSE error to this range, 2 times RMSE yields a 95% confidence interval
- Temperature ranges for all layers large
 - Due to uncertainty in contact conductance
 - Methods for reducing range:
 - Additional testing to better characterize outsole-to-regolith conductance
 - Refinement of model correlation with emphasis on specific temperatures
 - Narrowing confidence interval



Conclusions

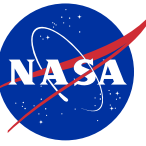
- Successfully correlated lunar boot thermal model using extensive, repeatable steady-state data obtained across 10 test points including contact and radiation cases
 - Achieved heat leak accuracy within 0.1 W
- Used bootstrapping and Monte Carlo methods to quantify model uncertainty for both test and operational conditions
- Outsole-to-regolith contact conductance identified as the dominant driver of heat flux and temperature variability
 - Bounded its impact and defined permissible environmental temperatures with parametric studies
- Future improvements needed in regolith contact fidelity, colder/more uniform radiation environments, and human in the loop testing to validate a high-fidelity metabolic, transient, foot model



- [1] E. Quick, "Model Correlation and Thermal Analysis of xEMU Boot at Lunar South Pole Temperatures," in *International Conference of Environmental Systems*, Rio Grande, Puerto Rico, 2026.
- [2] B. Swartout, "xEMU Spacesuit Boot Lunar South Pole Thermal Vacuum Testing," in *International Conference of Environmental Systems*, Rio Grande, Puerto Rico, 2026.
- [3] B. Swartout, "Exploration Extravehicular Mobility Unit (xEMU) Lunar Boot Chamber B Thermal Vacuum Testing Results," in *International Conference on Environmental Systems*, Louisville, 2024.
- [4] B. C. Conger, "Thermal Analysis of the EMU Boot with Foam Insert Removed," CTSD 1404, 1992.
- [5] T. Durrant, "Orlan Spacesuit Boot Thermal Model," HDID-2G41-1132, 1999.
- [6] J. V. Iovine, "EMU Boot Thermal Analysis for ISS EVA," HDID-2G41-954, 1998.
- [7] J. V. Iovine, "Thermal Analysis of the EMU Boot Pressure Bladder," CTSD862, 1990.
- [8] W. Lee, "Thermal Analysis Results of Stand-Alone Shuttle Boot Model for the Thermal Vacuum Chamber B Test for STS-37 Mission Hardware," CTSD454, 1989.
- [9] T. Nguyen, "The Thermal Effect of a Newly Designed Aluminum Piece Attached to the Heel of the EMU Boot," 20977, 1984.
- [10] D. Ochoa, "Thermal Design and Analysis of Constellation Space Suit System Boot Options for Lunar Missions," ESCG-4470-09-TEAN-DOC-0110, 2009.
- [11] D. G. Gilmore, *Spacecraft Thermal Control Handbook: Volume 1: Fundamental Technologies*, The Aerospace Press, 2002.
- [12] "fitgeotrans," Mathworks, [Online]. Available: <https://www.mathworks.com/help/images/ref/fitgeotrans.html>.
- [13] "imwarp," Mathworks, [Online]. Available: <https://www.mathworks.com/help/images/ref/imwarp.html>.
- [14] Z. Fester, S. McFarland, N. Andersen, E. Quick, D. Green, T. Jaimes and D. Morris, "xEMU Boot TVAC Final Report," NASA, CTSD-ADV-2272.
- [15] N. L. Langston and V. Margiott, "EMU Phase VI Glove Thermal Vacuum Test and Analysis Final Report," CTSD-SS-1621, 1998.
- [16] M. B. Abney, R. J. Jones, S. R. Wilson, T. K. Brady and N. L. Andersen, "Lunar Glove Thermal and Durability Analysis, Test, and Failure Mitigation," NESC-RP-21-01722, 2024.
- [17] NASA, "NASA Spaceflight Human-System Standard Volume 1, Crew Health," 2023.



Acknowledgements / Questions



The authors would like to acknowledge the CITADEL test team at JPL for their hard work conducting test points and compiling data files.

Thank you!

Questions?

Further inquiries may be sent to the following:

Emma Quick: emma.j.quick@nasa.gov

Noah Andersen: noah.l.andersen@nasa.gov