ADVANCED EXTRAVEHICULAR PROTECTIVE SYSTEMS
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INTRODUCTION
The U.S. manned space effort planned for the 1980s consists of long-duration missions with
orbiting space stations, potential lunar bases, and possibly Mars landings. Extravehicular activity
(EVA) is likely to take an increasingly important role in the completion of these future missions.
With a potential need for two or three EVA missions per man per week, however, the use of
expendables in the portable life support system may become prohibitively expensive and
burdensome. For future EVAs to be effective in the total systems context, the portable life support
system may need a regenerable capacity.

Hamilton Standard, with NASA funding, has been conducting an advanced extravehicular
protective system (AEPS) study program. The objective of this AEPS study is to provide a
meaningful appraisal of various regenerable and partially regenerable portable life support system
concepts for EVA use in the 1980s and to identify the required new technology area.

STUDY METHOD
Study Evaluation Criteria
Selection of the most favorable EVA subsystem and system equipment has always posed a difficult
problem. This is particularly true for the AEPS study, which deals with long-duration earth orbital,
lunar surface, and martian surface missions in which the vehicle penalty for an AEPS configuration
is more important than it was for the shorter term Gemini and Apollo programs. This factor reduces
the validity of the traditional heavy emphasis on EVA equipment equivalent volume and weight
within the evaluation criteria. Thus, to fulfill the objective of the AEPS study, it has been necessary
to establish criteria reflecting an objective evaluation of not only the EVA crewman and his
equipment, but also of the parent vehicle or shelter and the total mission.

The determination of the AEPS study selection criteria is based on a recognition
that some requirements are absolute, others are of primary importance, and still others
represent second-order effects. The criteria used as a basis for the AEPS subsystem and
system selection are shown in figure 7.1. The criteria are applied sequentially in the groups
shown to eliminate concepts that fail on either an absolute (go/no go) or comparative
basis, and to provide the basis for selection among surviving candidates.
The go/no go criteria define the minimum acceptable requirements for a concept. If a concept does not meet, or cannot be modified or augmented to meet, all the go/no go criteria, it receives no further consideration in the study and is listed as unacceptable and eliminated.

The primary criteria are the principal evaluation criteria for all concepts that pass the go/no go criteria requirements. The ratings applied to a candidate concept depend on its characteristics relative to the other candidates. Each candidate concept receives a rating of from 0 to 100 for each primary criterion. Each rating is then multiplied by the weighting factors defined in table 7.1 and these are added to obtain a total rating for each candidate concept. A candidate concept is selected if its overall rating is clearly the best of the competing concepts. If a clear-cut choice is not evident, the remaining competing concepts are reviewed against the secondary criteria.

Table 7.1 Primary criteria weighting factors.

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<thead>
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<th>Weighting Factors</th>
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<tr>
<td></td>
<td>space station</td>
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<tr>
<td>Vehicle equivalent weight</td>
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</tr>
<tr>
<td>AEPS equivalent volume</td>
<td>0.30</td>
</tr>
<tr>
<td>Reliability</td>
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<td>Operability</td>
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<tr>
<td>Flexibility</td>
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The secondary criteria represent a step in depth of competitive evaluation that is taken if no clear-cut selection is available from the primary criteria. Ratings of the candidate concepts against secondary characteristics are relative assessments within each area of consideration and, as in the implementation of the primary criteria, each candidate concept receives a rating from 0 to 100 for each criterion. Each rating is then multiplied by the weighting factors defined in table 7.2 and these are added to obtain a total rating for each candidate concept. A candidate concept is selected if its overall rating is clearly the best of all competing concepts.

In any event, the secondary criteria are applied against all recommended concepts to provide a systematic review of the overall acceptability of these selected concepts.

Study Flow
The AEPS study consisted of the following four basic tasks.

Study Plan and Specifications. The basic ingredients to a meaningful AEPS study are the study plan and the AEPS specifications. The diverse nature of earth orbital, lunar, and martian applications required that separate AEPS specifications be generated for each application.

The goal of these specifications are general guidelines representing the probable trends for earth orbital, lunar base, and martian landing missions in the 1980s. To support the specification generation effort, baseline EVA mission models were established to define work performance tasks and required crew skills; to determine representative time allocations for these tasks; to define
operational procedures for donning/doffing, checkout, egress/ingress, recharge/regeneration, etc.; and to define applicable interface areas. In addition, vehicle environmental control/life support system models were established to serve as guides to determine the AEPS recharge/regeneration capabilities of the vehicle.

**Subsystem Studies.** The first step in the subsystem definition study was the preparation of subsystem requirements for each major functional area of each configuration. Based on these requirements candidate concepts were identified in each of the major subsystem areas (carbon dioxide control/oxygen supply, trace contaminant control, thermal control/humidity control, and power). In areas where in-house data were not complete, a literature survey was conducted and industry contacts made, as required. Once all data were assembled and candidate subsystem concepts identified, a preliminary evaluation was conducted to screen out and reject the obviously noncompetitive candidates. Performance characteristics (such as flow rates, temperature levels and pressure levels) of the selected candidate subsystems were roughly determined and preliminary schematics and component lists generated. The candidate subsystems were then sized to meet the subsystem requirements.

These subsystems were then compared against the go/no go evaluation criteria. If a concept was found unacceptable, sufficient auxiliary equipment was added to that subsystem, if possible, to meet the go/no go criteria. If a candidate concept could not be made acceptable, it was removed from further consideration at that point.

A parametric analysis of the remaining candidate subsystem concepts was then conducted. The following data were generated as required for comparison purposes among the candidate subsystems.

1. Vehicle launch weight (including expendables, spares, recharge or regeneration equipment, and checkout equipment, in addition to the basic subsystem) versus total mission duration.
2. EVA equipment volume versus EVA mission duration.
4. EVA equipment weight versus EVA mission duration.

The remaining candidate subsystems were then compared against the primary criteria. Further equipment was added or the arrangements modified, as required, to upgrade candidate subsystem concepts found unacceptable or inferior relative to the reliability and operability criteria. Of course, the associated weight, volume, and power penalties were also reflected in the parametric analyses. If

### Table 7.2 Secondary criteria weighting factors.

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<td>Vehicle equivalent volume</td>
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<td>AEPS equivalent weight</td>
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<tr>
<td>Interface compatibility</td>
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<td>Cost</td>
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a candidate concept could not be made acceptable, or was still obviously grossly inferior to the other candidates, it was removed from further consideration at that point.

If a clear-cut choice still could not be made from the primary criteria evaluation, the remaining candidate subsystem concepts were compared against the secondary criteria. As in the primary criteria evaluation, equipment modifications were made to a candidate concept(s) to upgrade it relative to the qualitative criteria (interface compatibility and maintainability) if it appeared inferior to other competing concepts. Again, the associated penalties were reflected in the parametric analyses.

From the results of the subsystem evaluations, the best competing subsystems were selected for each AEPS configuration. Several subsystems that perform the same function were recommended for further study on the system levels.

**System Studies.** After completion of the subsystem studies, a systems integration effort was conducted wherein the selected candidate subsystem concepts were combined into several candidate baseline space station, lunar base, and martian base AEPS systems. The systems integration effort evaluated and defined the following elements, which could not be fully evaluated on the subsystem level:

1. Subsystem interfaces (both functional and physical).
2. Instrumentation and controls.
3. Thermal balance.
4. Equipment power requirements.
5. Humidity control.
6. Method of heat transport to heat rejection system and associated coolant flows.
7. Trace contaminant requirements.
8. Suit and vehicle interfaces.

The candidate baseline systems were then subjected to a competitive evaluation utilizing the established criteria and the parametric results of the subsystem studies. Results of the systems evaluation led to the selection of the candidate space station, lunar base and martian base AEPS baseline concepts.

Prior to a final review and iteration of the AEPS baseline concepts, Hamilton Standard reviewed with NASA the general specifications and the evaluation criteria to assure that both are still consistent with the objectives of the study and the results to date. Upon satisfactory completion of this task, the AEPS baseline concepts were subjected to a detailed performance review and evaluation to optimize system performance. Components and subsystems were resized and system arrangements modified, as required.

The operational modes of each baseline concept were reviewed in detail to simplify operational procedures. Specific emphasis was placed on:

1. Startup, checkout, and shutdown procedures.
2. Recharge and regeneration procedures.

A safety/reliability evaluation of each baseline concept was conducted. Top level system FMEAs were conducted and single point and sequential failures were eliminated. This analysis also formed the basis for selection of the AEPS instrumentation.

The interface compatibility of each AEPS baseline concept was evaluated with respect to the crew, the space suit, the vehicle, and other EVA equipment. Specific emphasis was placed on location of AEPS control and displays, use of the time independent module/time dependent module.
(TIM/TDM) concept, partial and full integration of the AEPS into the space suit, and compatibility of the AEPS subsystems with vehicle EC/LSS subsystems.

The final AEPS system recommendations resulted from this total effort.

**New Technology.** After establishment of the AEPS baseline concepts, a portion of the study effort was directed toward generation of a priority listing of new technology development activity required to permit the AEPS recommendations to be implemented.

The principal objectives of this effort were:

1. To provide confirmation of attractive concepts where, although feasibility may have been demonstrated, development status and confidence is marginal.
2. To define problems, recommend approaches, and estimate resources to solve these problems.

**SUBSYSTEM STUDIES**

To ensure that the results of this study were both meaningful and useful for future related efforts, Hamilton Standard adopted a broad-based approach to candidate subsystem concept identification. The whole gamut of concept approaches was investigated with a specific effort on our part to preclude any prejudgment of concept value prior to concept identification. Specific emphasis was placed in the areas of thermal control and CO₂ control/O₂ supply, as they represented the areas where the greatest benefits could be derived through reduction of vehicle penalties and AEPS volume and weight.

Initial effort results in the identification of 55 candidate thermal control concepts (table 7.3), 21 candidate CO₂ control concepts (table 7.4), 14 candidate O₂ supply concepts (table 7.5), and 3 candidate O₂ generation concepts (table 7.6). All these concepts were evaluated on a cursory basis, and those that were deemed to be “obviously noncompetitive” were eliminated. Of these original candidate concepts identified and analyzed on a preliminary basis, 25 thermal control concepts and 19 combined CO₂ control/O₂ supply concepts were carried into the evaluation. These candidate concepts were subjected to the go/no go, primary, and secondary evaluations in consecutive order and in accordance with the procedure described earlier.

**Table 7.3 Thermal control concepts.**

*Expendables*

- **Water**
  - Water boiler
  - Supercooled water boiler
  - Supercooled water boiler with vapor regenerative cooling
  - Water sublimator
  - Supercooled water sublimator
  - Supercooled water sublimator with vapor regenerative cooling
  - Plate fin flash evaporator
  - Nonsteady state pulse feed flash evaporator
  - Static vortex flash evaporator
  - Turbine-rotary vortex flash evaporator
Table 7.3 *Thermal control concepts.* (Continued)

**Expendables (continued)**

Water (continued)

- Motor-rotary vortex flash evaporator
- Multistage flash evaporator
- Vapor diffusion through suit pressure valves
- Vapor diffusion through water permeable membrane

Hydrogen Peroxide (H₂O₂)

- H₂O₂ dissociation into H₂O and O₂

Ammonia (NH₃)

- NH₃ boiler
- NH₃ sublimator

Carbon Dioxide (CO₂)

- CO₂ boiler
- CO₂ sublimator

Methane (CH₄)

- CH₄ sublimator

Cryogenics

- Cryogenic O₂
- Cryogenic H₂

Feces/urine sludge

- Evaporation of H₂O from feces/urine sludge

**Conduction**

Conduction via the lunar or martian surface

**Convection (Mars only)**

- Free convection
- Forced convection
- Hilsch tube
Table 7.3  Thermal control concepts. (Concluded)

Radiation

Direct cooling
LCG
Heat pipe
Water adsorption using:
- LiCl • 3H₂O
- CaCl • 6H₂O
- Molecular sieve
- Silica gel
- LiBr • 3H₂O
- Na₂Se • 16H₂O

Indirect cooling
- Vapor compression refrigeration cycle using Freon
- Water adsorption cycle using NH₃
- Water adsorption cycle using LiBr
- Brayton cycle using Air

Thermal Storage

Ice
Subcooled ice
- Thermal wax—transit 86
- Eutectic salt—sodium sulphate (Na₂SO₄ • 10H₂O)
- Phosphonium chloride (PH₄C1)
- Hydrogen (H₂)
- Lunar or martian rock

Energy Conversion

Thermoelectric
Thermionic
Thermodielectric

Hybrids
- Expendable/radiation—direct cooling
- Expendable/radiation—indirect cooling
- Expendable/thermal storage
- Radiation/thermal storage
- Thermal storage/water adsorption
Table 7.4  Carbon dioxide control concepts.

Expendables

Solid sorbents
   Hydroxides
   Superoxides
   Peroxides
   Ozonides

Liquid sorbent
   Hydroxide solutions

Open loop
   Purge flow

Regenerables

Solid sorbents
   Activated charcoal
   Molecular sieve
   Metallic oxides
   Solid amines

Liquid sorbents
   Carbonate solutions
   Liquid amines

Electrochemical
   Hydrogen depolarized cell
   Two-stage carbonation cell
   One-stage carbonation cell
   Electrodialysis
   Fused salt

Mechanical

   Simple membrane diffusion
   Immobilized liquid membrane diffusion
   Mechanical freezeout
   Cryogenic freezeout

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Table 7.5 Oxygen supply concepts.

\[ O_2 \text{ Storage} \]
- Gaseous
- Supercritical utilizing thermal pressurization
- Subcritical utilizing thermal pressurization
- Subcritical utilizing positive expulsion
- Solid

\[ \text{Solid Decomposition} \]
- Superoxides
- Peroxides
- Ozonides
- Sodium chlorate candles (NaClO\textsubscript{3})
- Lithium perchlorate candles (LiClO\textsubscript{4})

\[ \text{Liquid Decomposition} \]
- Hydrogen peroxide (H\textsubscript{2}O\textsubscript{2})
- Reactant storage (N\textsubscript{2}H\textsubscript{4}/N\textsubscript{2}O\textsubscript{4})
- Reactant storage (N\textsubscript{2}H\textsubscript{4}/H\textsubscript{2}O\textsubscript{2})

\[ \text{Electrolysis} \]
- Water electrolysis

Table 7.6 Oxygen generation concepts.

- Solid electrolyte
- Bosch reactor/water electrolysis
- Sabatier reactor/water electrolysis
Thermal Control

As a result of these evaluations, the following three general thermal control categories were selected for further study:

1. Expendable concepts utilizing water.
2. Radiation.
3. Thermal storage.

Five specific thermal control subsystem concepts were recommended to be carried into the systems integration phase of the AEPS study.

Water Boiler. The water boiler (fig. 7.2) is an expendable thermal control concept that utilizes the heat of vaporization of water to provide direct cooling of the liquid cooling garment (LCG) loop and vent loop. The wick-fed water boiler also acts as the storage vessel for the expendable water. The expendable water boiling temperature is controlled by a back pressure valve, which is either a temperature sensing or pressure sensing flow control valve. Crewman comfort is achieved automatically by the temperature control valve. Water removed by the water separator is fed into the water boiler, thus providing additional cooling capacity. A relief valve furnishes protection against overpressurization due to storage temperature fluctuations. Recharge is simply accomplished utilizing the fill valve. The water boiler is being recommended as a representative concept of all the expendable water concepts that utilize phase change to reject heat.

Thermal Storage with Phosphonium Chloride (PH₄Cl). Thermal storage (fig. 7.3) utilizing phosphonium chloride is a self-regenerable thermal control concept. The heat of fusion of PH₄Cl is 324 Btu/lb at 82°F and above 48 atm pressure. A vapor compression intermediate loop is utilized to raise the desired coolant temperature of 50°F at the vent loop/liquid heat transport loop heat exchanger to 82°F at the thermal storage unit. Humidity control is furnished by a water separator and holding tank that remove and store vent loop condensate. Vehicle penalties associated with this concept are relatively low since PH₄Cl will resolidify of its own accord at normal cabin temperatures.

Solid PH₄Cl sublimes at pressures below 500 psia at room temperature. As pressure is decreased further, gaseous PH₄Cl dissociates into two gases: hydrogen chloride and phosphene.
Expendable/Direct Radiative Cooling. This concept (fig. 7.4) is a hybrid consisting of a water boiler and radiator connected in parallel through the LCG temperature control valve. The temperature control valve selects the percentage of the heat load from the LCG that is shared by each subsystem. The radiator is sized to handle the average heat load while the water boiler handles peak loads; thus radiator size and water expended in the boiler are minimized. Humidity control is provided by a condensing heat exchanger and a water separator that feeds the separated water to the water boiler to provide additional cooling capacity. For low or no-load conditions, a variable conductance, area, or emissivity device must be used to prevent overcooling of the LCG.

Expendable/Heat Pump. This hybrid concept (fig. 7.5) consists of a radiator/vapor compression cycle and a water boiler connected in parallel through an automatic LCG temperature control valve. The temperature control valve selects the percentage of the heat load from the LCG that is shared by each subsystem. The radiator/vapor compression subsystem is sized to handle the average LCG heat load plus the heat load from the vent system while the water boiler handles peak heat loads. This minimizes radiator size, compressor size, and power consumption as well as water expended in the boiler. Humidity control is provided by the vapor compression cycle evaporator and the water separator that feeds the separated water to the water boiler to provide additional cooling capacity. For low or no-load conditions, a radiator bypass or compressor short circuit (not shown in schematic) or variable speed compressor is required to prevent overcooling at the evaporator.

Expendable/Thermal Storage (PH₄Cl). This hybrid concept (fig. 7.6) utilizes a water boiler in parallel with a PH₄Cl thermal storage unit via an LCG temperature control valve. The temperature control valve selects the percentage of the heat load from the LCG that is shared by each subsystem, the intention being that the PH₄Cl thermal storage unit will handle the average heat load and the
of solid regenerable sorbents were identified as candidate materials: metallic oxides and solid amines.

The CO₂ control/O₂ supply subsystem concepts recommended to be carried into the systems integration phase of the AEPS study are described below. All of the concepts utilize a high-pressure gaseous O₂ supply system.

**Metallic Oxides - Vehicle Regenerable.** Metallic oxides (ZnO, MgO) react with CO₂ according to the reversible reaction

\[
MO + CO₂ \rightleftharpoons MCO₃ + Q
\]

The carbonate readily decomposes with increasing temperature and, in some cases, may be solely vacuum regenerable. However, excessive volume change during the adsorb/desorb cycle affects the chemical's physical stability and is a prime consideration in any future development effort. For this study, the adsorbent was contained between screens with gas flow over rather than through the packaging. CO₂ diffusion into the thin oxide bed will be sufficient as long as the solid volume transition during adsorb/desorb does not result in an impregnable surface or if any extremely fine screen is not required. An alternate concept would consider a carrier to stabilize the solid adsorbent—possibly a thin layer of the oxide flame sprayed on a screen matrix.

In the vehicle regenerable configuration (fig. 7.7), the adsorbent is packaged in a cartridge that is replaced after each mission. An oven/vacuum chamber is provided within the vehicle for cartridge regeneration. Reclamation of the oxygen is possible with this system by directing gas to the vehicle CO₂ reduction system.

**Metallic Oxides - AEPS Regenerable.** A variation of the metallic oxide concept considers a cyclic or AEPS regenerable configuration (fig. 7.8). This concept provides for regeneration of the metallic oxide subsystem during the actual EVA mission. Two beds, similar in design to that described for the vehicle regenerable system, are provided, each containing electrical elements for regeneration and a cooling loop to cool the regenerated bed and maintain temperature control during operation.

water boiler will handle peak loads. Compressor power and expendable water thus are minimized. The water boiler provides humidity control by cooling the vent loop that feeds the separated water to the boiler via the water separator to provide additional cooling capacity. A bypass valve is utilized in the thermal storage subsystem to prevent overcooling under low or no-load conditions. This system is flexible in that it can be sized for a multitude of thermal load sharing combinations.

**Carbon Dioxide Control/Oxygen Supply**

One general CO₂ control/O₂ supply category was selected for further study—a solid regenerable sorbent combined with a high-pressure gaseous oxygen supply system. Two families
A timer is provided to sequence the vent loop and coolant loop valves to allow the vent loop and coolant loop to flow to the onstream bed and to heat and expose to space vacuum the regenerating bed.

**Solid Amine - AEPS Regenerable.** An inert carrier is utilized to provide a stable amine adsorbent bed in this concept (fig. 7.9). The regenerable solid amine is packaged within the flow passages of a plate-fin matrix similar in design to an extended surface compact heat exchanger. Alternate flow passages contain adsorbing and desorbing material with the unique feature of an isothermal process. Energy released from the adsorbing passages is transferred by conduction through the metal matrix to the desorbing material to supply the requirements of the endothermic desorption. This concept neither imposes a thermal load on the AEPS nor requires energy for regeneration. A timer and valving is provided to cycle the packed beds from the online adsorb to the space vacuum desorb cycle.

The parametric analyses of the recommended subsystem concepts were then reviewed and updated to reflect the latest AEPS specification requirements for each of the three missions, space station, lunar base, and martian. Parametric data generated includes:

1. Vehicle equivalent weight versus total mission duration.
2. Vehicle equivalent volume versus total mission duration.
3. AEPS equivalent volume versus EVA mission duration.
4. AEPS equivalent weight versus EVA mission duration.
5. Accumulated resupply launch weight versus number of resupplies.
The results of these parametric analyses, which were utilized for selecting and sizing subsystems for the system studies, are presented in the following figures:

**Figure 7.10 (a) Thermal control for space station AEPS.**

**Figure 7.10 (b) Thermal control for space station AEPS.**
Figure 7.10 (c) Thermal control for space station AEPS.

Figure 7.10 (d) Thermal control for space station AEPS.
Figure 7.10 (e) Thermal control for space station AEPS.

Figure 7.11 (a) Thermal control for lunar base AEPS.
Figure 7.11 (b) Thermal control for lunar base AEPS.

Figure 7.11 (c) Thermal control for lunar base AEPS.
Figure 7.11 (d) Thermal control for lunar base AEPS.

Figure 7.11 (e) Thermal control for lunar base AEPS.
Figure 7.12 (a) Thermal control for martian AEPS.

Figure 7.12 (b) Thermal control for martian AEPS.
Figure 7.12 (c) Thermal control for martian AEPS.

Figure 7.12 (d) Thermal control for martian AEPS.
Figure 7.13 (a) Carbon dioxide control/oxygen supply for space station AEPS.

Figure 7.13 (b) Carbon dioxide control/oxygen supply for space station AEPS.
Figure 7.13 (c) Carbon dioxide control/oxygen supply for space station AEPS.

Figure 7.13 (d) Carbon dioxide control/oxygen supply for space station AEPS.
Figure 7.13 (e) Carbon dioxide control/oxygen supply for space station AEPS.

Figure 7.14 (a) Carbon dioxide control/oxygen supply for lunar base AEPS.
Figure 7.14 (b) Carbon dioxide control/oxygen supply for lunar base AEPS.

Figure 7.14 (c) Carbon dioxide control/oxygen supply for lunar base AEPS.
Figure 7.14 (d)  *Carbon dioxide control/oxygen supply for lunar base AEPS.*

Figure 7.14 (e)  *Carbon dioxide control/oxygen supply for lunar base AEPS.*
Figure 7.15 (a) Carbon dioxide control/oxygen supply for martian AEPS.

Figure 7.15 (b) Carbon dioxide control/oxygen supply for martian AEPS.
Figure 7.15 (c) Carbon dioxide control/oxygen supply for martian AEPS.

Figure 7.15 (d) Carbon dioxide control/oxygen supply for martian AEPS.
The system studies combined the selected candidate subsystem concepts into candidate baseline space station, lunar base, and martian AEPS schematics. The following schematics and flow charts are representative of potential AEPS configurations that might result if the technology recommendations emanating from the AEPS study are implemented.

### AEPS Concepts

**Space Station 1 (fig. 7.16).** This AEPS concept contains all required life support equipment for extravehicular operation including an O₂ ventilation loop, a high-pressure O₂ subsystem, a water heat transport loop, a Freon 12 heat transport loop, a power supply, instrumentation, and operating controls and displays. The O₂ ventilation loop circulates a reconditioned and replenished O₂ supply through the suit. The O₂ from the suit enters the atmosphere regeneration subsystem and first passes through the debris trap where solid particles and droplets are removed; next CO₂ is removed by both physical adsorption and chemical absorption using a vehicle regenerable metallic oxide—zinc oxide; odors and trace contaminants are removed through physical adsorption by the activated charcoal in the contaminant control canister; and finally, an absolute filter provides dust and bacteria control. The O₂ then passes to a Freon evaporator heat exchanger that cools the circulated O₂ and condenses the entrained moisture. The cooled O₂ continues to the water separator where the condensed water vapor is removed and transferred to the water boiler to provide additional cooling capacity. The cool, dry O₂ then passes to the fan that circulates a ventilation flow of 6 acfm to the suit.

The high-pressure O₂ subsystem contains 0.75 lb of usable O₂ at 6000 psia and 65 °F, and regulates the pressure in the O₂ ventilation loop to 7.0 ± 0.1 psia. This subsystem consists of an O₂ bottle, fill fitting, pressure sensor, shutoff valve, and pressure regulator.

The water heat transport loop cools the suited crewman by supplying and circulating cool water through a network of tubes built into his undergarment. The skin is cooled by direct conduction, and the mean skin temperature is lowered to a level where little, if any, perspiration occurs. A pump circulates the cooled water through the water heat transport loop at a flow rate of 4 lb/min. Flow through the thermal control subsystem is regulated by an automatic temperature control valve.

The thermal control subsystem is a hybrid expendable/thermal storage concept that consists of a water boiler and phosphonium chloride thermal storage unit. A Freon 12 heat transport loop consisting of a Freon evaporator, a variable speed compressor, and a variable orifice is utilized to transfer heat added at the evaporator to the phase change thermal storage unit. Average thermal loads are handled by the Freon evaporator. However, as the heat load increases above average levels, an increasing quantity of flow is precooled in the water boiler to prevent overloading of the evaporator and resultant loss of humidity control.

The estimated total volume and weight for this AEPS configuration are 2500 in.³ and 77 lb based on average metabolic rate of 1000 Btu/hr for an EVA duration of 4 hr.

**Space Station 2 (fig. 7.17).** This AEPS concept is similar to concept 1 except:

1. The vehicle regenerable metallic oxide CO₂ control subsystem is replaced by an AEPS regenerable solid amine plate-fin matrix that removes both CO₂ and water vapor from the O₂ ventilation loop thus providing both CO₂ and humidity control. This is a cyclic concept using a 30-min full cycle. Energy released during the adsorb cycle is conducted to the regeneration portion of the subsystem thus supplying the endothermic heat of desorption.
Figure 7.16 AEPS concept 1, space station.
### Station

#### Liquid Loop

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#### Freon Loop

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<tr>
<td>Pressure, psia</td>
<td>61.4</td>
<td>103</td>
<td>102</td>
<td>61.8</td>
</tr>
</tbody>
</table>

**Figure 7.16** AEPS concept 1, space station. (Concluded)

---

![Diagram](image)

**Figure 7.17** AEPS concept 2, space station.
The thermal control subsystem is a hybrid expendable/radiation heat pump subsystem and consists of a water boiler and a Freon 12 refrigeration system. The Freon 12 refrigeration system consists of a Freon evaporator, a variable speed compressor, a high temperature radiator, and a variable orifice. The Freon system is sized to reject average heat loads at nighttime conditions. Heat in excess of this amount is rejected by the water boiler. The automatic temperature control valve maintains the correct flow split between the two thermal control subsystems as well as conditioning the liquid transport loop. The estimated total volume and weight for this AEPS configuration are 3100 cu in. and 65 lb based on an average metabolic rate of 1000 Btu/hr for an EVA duration of 4 hr.

Lunar Base 3 (fig. 7.18). This AEPS concept is similar to concept 2 except the thermal control subsystem is composed of a PH4Cl thermal storage unit and a Freon 12 refrigeration cycle consisting of a Freon evaporator, a variable speed compressor, and a variable orifice. Heat is added at the evaporator and stored at the thermal storage unit by the melting of PH4Cl. The estimated total volume and weight for this AEPS configuration is 4500 cu in. and 193 lb based on an average metabolic rate of 1050 Btu/hr for an EVA duration of 8 hr.

### Vent Loop

<table>
<thead>
<tr>
<th>Temperature, °F</th>
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<tr>
<td>Volume flow rate, FT³/min</td>
<td>6.315</td>
<td>6.32</td>
<td>6.18</td>
<td>6.22</td>
<td>5.92</td>
<td>5.97</td>
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<tr>
<td>Total pressure, psia</td>
<td>6.9</td>
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<td>6.987</td>
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<td>O₂ weight flow, lb/hr</td>
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<td>13.531</td>
<td>13.531</td>
<td>13.531</td>
<td>13.531</td>
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<tr>
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<td>0.337</td>
<td>0.205</td>
<td>0.205</td>
<td>0.205</td>
<td>0.205</td>
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### Liquid Loop

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### Freon Loop

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Figure 7.17 AEPS concept 2, space station. (Concluded)
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<td>13.52</td>
<td>13.52</td>
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<td>.367</td>
<td>.162</td>
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<td>.337</td>
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Figure 7.18 AEPS concept 3, lunar base.
### Liquid Loop

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<td>60</td>
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### Freon Loop

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<td>Pressure, psia</td>
<td>61.4</td>
<td>103</td>
<td>102</td>
<td>61.8</td>
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---

**Figure 7.18** AEPS concept 3, lunar base. (Concluded)

**Lunar Base 4 (fig. 7.19).** This AEPS concept is similar to concept 1 except the thermal control subsystem utilized is similar to that described in concept 2. The estimated total volume and weight for this AEPS configuration is 3500 cu in. and 100 lb based on an average metabolic rate of 1050 Btu/hr for an EVA duration of 8 hr.

---

**Figure 7.19** AEPS concept 4, lunar base.
### Vent Loop

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<td>72</td>
</tr>
<tr>
<td>Volume flow rate, FT³/min</td>
<td>6.3</td>
<td>6.3</td>
<td>6.25</td>
<td>6.29</td>
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<td>5.93</td>
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<td>O₂ weight flow, lb/hr</td>
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<td>13.52</td>
<td>13.52</td>
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<td>13.52</td>
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<td>0.178</td>
</tr>
<tr>
<td>CO₂ weight flow, lb/hr</td>
<td>.367</td>
<td>.367</td>
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</tr>
<tr>
<td>H₂O weight flow, lb/hr</td>
<td>.337</td>
<td>.337</td>
<td>.337</td>
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<td>.205</td>
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### Liquid Loop

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### Freon Loop

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</table>

**Figure 7.19**  *AEPS concept 4, lunar base. (Concluded)*

**Martian Base 5 (fig. 7.20)**  *This AEPS concept is similar to concept 1 except the thermal control subsystem is a hybrid expendable/direct radiative cooling subsystem and consists of a water boiler and a direct radiator. Water in the heat transport loop exiting the radiator is the coolant in the humidity control condensing heat exchanger. As the radiator load increases and exceeds design values, a portion of the water heat transport loop flow bypasses to the water boiler, thus maintaining a constant radiator outlet temperature. The estimated total volume and weight for this AEPS configuration is 3100 cu in. and 84 lb based on an average metabolic rate of 1200 Btu/hr for an EVA duration of 8 hr.*

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Figure 7.20 AEPS concept 5, martian base.
NEW TECHNOLOGY RECOMMENDATIONS

As a result of the AEPS Study Program, the following areas of required new technology were identified and are recommended for future research and development.

   Thermal Control

**Thermal Storage.** Investigate and develop a thermal storage material(s) whose heat of fusion exceeds 300 Btu/lb. One such candidate material, PH4Cl, has already been identified and analytically evaluated during the AEPS study.

**Radiation.** Investigate and develop radiator surface coatings and treatments to optimize performance and minimize potential surface degradation. In addition, develop a lightweight, deployable radiator concept.

   CO₂ Control

Develop a solid regenerable CO₂ sorbent that provides the performance, regeneration and life characteristics required for AEPS type applications. Two candidate families of solid regenerable sorbents—metallic oxides and solid amines—have already been identified and evaluated during the AEPS study.

   O₂ Supply

Develop a high cyclic life/high static strength material for a 6000 psi oxygen bottle.

   Power Supply

Operationally develop a high energy density, rechargeable electric storage battery. One candidate—a lithium-nickel halide battery—was identified during conduct of the AEPS study, and it (together with any other battery demonstrating a similar or greater energy capacity) is recommended for further research and development.

   Contaminant Control

Confirm or modify the AEPS contaminant model selected and determine the effect of long term intermittent exposure upon the suited crewman; then design, develop and test the contaminant control subsystem to confirm performance characteristics.

   Humidity Control

Results of the AEPS study indicate that a condensing heat exchanger in series with either an elbow wick separator or a hydrophobic/hydrophyllic screen separator are the optimum choices for an AEPS-type application. Research and development to determine the effect of contamination and bacterial/fungus growth upon the performance of both of these concepts is recommended to permit design and development of a long life humidity control subsystem.

   Prime Movers

Design and develop longer life prime movers (i.e., fan, pump, and variable-spaced compressor) that have higher compressor efficiency and lower electronics and bearing losses than those presently being utilized in aerospace programs.

   Automatic Temperature Control

Design and develop an automatic temperature controller. Further research and development is recommended to determine the signal parameters that provide accurate automatic temperature control, and to develop the required hardware.
Miscellaneous

a. Develop automated equipment to permit simple, rapid checkout of the AEPS. The present Apollo EMU PLSS requires approximately thirty (30) minutes for checkout prior to egress of the vehicle.

b. Investigate and evaluate potential integration (both functional and physical) of the crewman’s personal maneuvering equipment with the AEPS for EVA missions in a zero gravity environment.

c. Improve the thermal isolation characteristics of the Thermal Meteoroid Garment (TMG), thus decreasing the peak thermal load on the AEPS thermal control subsystem.

d. Improve the Liquid Cooling Garment (LCG) heat transfer characteristics. This permits the liquid heat transport loop to operate at a higher temperature and thus decreases the power penalty associated with the thermal control subsystems which utilize a vapor compression (heat pump) cycle.

e. Conduct manned testing to evaluate the short-term and long-term physiological effects of various candidate pressure suit levels (3.5 to 14.7 psia) upon the crewman. Specific factors to be determined are:
   - Required versus tolerable O₂ prebreathing time
   - O₂ partial pressure exposure limitations including frequency and duration
   - Safe decompression/recompression levels, rates and frequency