EFFECT OF INCREASED FUEL TEMPERATURE
ON EMISSIONS OF OXIDES OF NITROGEN
FROM A GAS TURBINE COMBUSTOR
BURNING ASTM JET-A FUEL

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SUMMARY

An annular gas turbine combustor was tested with heated ASTM Jet A fuel to determine the effect of increased fuel temperature on the formation of oxides of nitrogen ($NO_x$). Fuel temperature ranged from ambient to 700 K (800° F). Combustor pressure was 4 atmospheres, inlet-air temperature was 938 K (1050° F), and reference Mach number was 0.075. The $NO_x$ emission index increased at a rate of 6 percent per 100 K (180° F) increase in fuel temperature. The result agrees with the results obtained in similar tests with heated natural-gas fuel. The result is also in agreement with the result expected from a simplified combustion model which relates the increase in the formation rate of nitric oxide to the increase in stoichiometric flame temperature.

INTRODUCTION

This report presents the results of combustor tests which were conducted to determine the effect of increasing fuel temperature on the formation of oxides of nitrogen ($NO_x$) from liquid ASTM Jet-A fuel.

Proposed use of the aircraft fuel as a heat sink for supersonic flight would increase the fuel temperature. Since the formation rate of nitric oxide (NO) is strongly dependent on flame temperature, the increased fuel temperature may raise the flame temperature enough to significantly increase NO emissions. On the other hand, the flammability limits of the fuel increase with increasing fuel temperature, which may allow the fuel to be burned at off-stoichiometric fuel-air ratios, where the rate of formation of NO is lower. Reference 1 reports the effects of fuel temperature for natural-gas fuel. It was found that the $NO_x$ emission index increased by approximately 4 percent per 100 K
(180° F) increase in fuel temperature at combustor temperatures simulating that of supersonic cruise.

The investigation reported herein was conducted to determine how the inlet temperature of ASTM Jet A-fuel affects the NOX emission index and to compare these results with those of reference 1 for natural-gas fuel. The investigation was conducted with the same full annular ram induction combustor used in reference 1. Fuel temperatures were varied from ambient to 700 K (800° F) at combustor inlet conditions simulating supersonic cruise pressure and temperature of 4 atmospheres and 838 K (1050° F), respectively.

The units for physical quantities in this report are given in both the International System of Units (SI) and the U.S. customary system. However, measurements during the investigation were made in the U.S. customary system.

APPARATUS AND PROCEDURE

Facility

Testing was conducted in a closed-duct test facility of the Engine Components Research Laboratory of the Lewis Research Center. A schematic of this facility is shown in figure 1. A detailed description of the facility and instrumentation are contained in reference 2. All fluid flow rates and pressures are controlled remotely.

Test Combustor

The combustor tested was designed using the ram-induction approach and is described in reference 3. With this approach, the compressor discharge air is diffused less than it is in conventional combustors. The relatively high-velocity air is captured by scoops in the combustor liner and turned into the combustion and mixing zones. Vanes are used in the scoops to reduce pressure loss caused by the high-velocity turns. The high velocity and the steep angle of the entering air jets promote rapid mixing of both the fuel and air in the combustion zone and the burned gases and air in the dilution zone. The potential result of rapid mixing is a shorter combustor or, alternatively, a better exit temperature profile in the same length.

A cross section of the combustor is shown in figure 2. The outer diameter is approximately 1.06 meters (42 in.), and the length from compressor exit to turbine inlet is approximately 0.76 meter (30 in.). A snout on the combustor divides the diffuser into three concentric annular passages. The central passage conducts air to the combustor
headplates and the inner and outer passages supply air to the combustor liners. There are five rows of scoops on each of the inner and outer liners to turn the air into the combustion and dilution zones.

The snout and the combustor liners are shown in figure 3. Figure 3(a) is a view looking upstream into the combustor liner. The scoops in the inner and outer liners can be seen, as well as the openings in the headplate for the fuel nozzles and swirlers. Figure 3(b) is a view of the snout and the upstream end of the combustor liner. The V-shaped cutouts in the snout fit around struts in the diffuser. The circular holes through the snout walls are for the fuel nozzle struts. Figure 3(c) gives a closer view of the liner and headplate, showing the liquid fuel nozzles and swirlers in place. There are a total of 24 fuel nozzles in the combustor.

Exhaust-Gas Sampling

Concentrations of nitric oxide, total oxides of nitrogen, carbon monoxide, unburned hydrocarbons, and carbon dioxide were obtained with an on-line system. The samples were drawn at the combustor exit from three circumferential locations (120° apart) and at five radial positions, through water-cooled stainless-steel probes. The exit instrumentation plane is shown in figure 2. The sample probe is pictured in figure 4.

Gas sampling system. - The samples collected by the three sampling probes were common manifolded to one sampling line. Approximately 18 meters (60 ft) of 0.95-centimeter (3/8-in.) stainless-steel line was used to transport the sample to the analytical instruments. To prevent condensation of water and to minimize adsorption-desorption effects of hydrocarbon compounds, the line was electrically heated to 420 K (310° F). Sampling line pressure was maintained at 1.7 atmospheres absolute to supply sufficient pressure to operate the instruments. Sufficient sample was vented at the instruments to provide a line residence time of about 2 seconds.

The exhaust-gas analysis system shown in figure 5 is a packaged unit consisting of four commercially available instruments along with associated peripheral equipment necessary for sample conditioning and instrument calibration. In addition to visual readout, electrical inputs were provided to an IBM 360 computer for on-line analysis and evaluation of the data.

The hydrocarbon content of the exhaust gas was determined by a Beckman Instruments Model 402 Hydrocarbon Analyzer. This instrument is of the flame ionization detector type.

The concentration of the oxides of nitrogen was determined by a Thermo Electron Corporation Model 10A Chemiluminescent Analyzer. The instrument includes a thermal converter to reduce NO\textsubscript{2} to NO and was operated at 973 K (1290° F).
Both carbon monoxide and carbon dioxide analyzers are of the nondispersive infrared (NDIR)-type (Beckman Instruments Model 315B). The CO analyzer has four ranges: 0 to 100 ppm, 0 to 1000 ppm, 0 to 1 percent, and 0 to 10 percent. This range of sensitivity is accomplished by using stacked cells of 0.65-centimeter (0.25-in.) and 34-centimeter (13.5-in.) length. The CO₂ analyzer has two ranges, 0 to 5 percent and 0 to 10 percent, with a sample cell length of 0.32 centimeter (0.125 in.).

Analytical procedure. - All analyzers were checked for zero and span prior to the test. Solenoid switching within the console allows rapid selection of zero, span, or sample modes. Therefore, it was possible to perform frequent checks to ensure calibration accuracy without disrupting testing.

Where appropriate, the measured quantities were corrected for water vapor removed. The correction included inlet-air humidity and water vapor from combustion. The equations used were obtained from reference 4.

The emission levels of all the constituents were converted to an emission index (EI) parameter. The EI may be computed from the measured quantities as proposed in reference 4 or by an alternate procedure which uses the metered fuel-air ratio when this is accurately known. With the latter scheme the EI for any constituent X is given by

\[
EI_X = \frac{m_X}{m_e} \frac{1 + f}{f} \frac{[X]}{10^{-3}}
\]  

(1)

where

- \(EI_X\) emission index in grams of \(X\) per kg of fuel burned
- \(m_X\) molecular weight of \(X\)
- \(m_e\) average molecular weight of exhaust gas
- \(f\) metered fuel-air ratio
- \(X\) measured concentration of \(X\) in ppm

Both procedures yield identical results when the sample validity is good.

Fuel System

The fuel system shown schematically in figure 1 was limited in fuel heating capability. The fuel heater was designed to heat 0.45 kg/sec (1 lb/sec) of natural-gas fuel to approximately 800 K (980° F) for the work done in reference 1. The fuel heater had to be converted for use with Jet-A fuel because it requires more enthalpy than natural
gas to arrive at the same temperature. Table I lists the enthalpy properties of Jet-A (ref. 5) and natural-gas fuels (ref. 6).

A set of available fuel nozzles was chosen which would maintain a high back pressure in the fuel heater and fuel manifold system. The high back pressure (>18 atm) was required to heat the fuel in one phase and thus avoid fractionation and coking in the fuel heater. Also, once the fuel is heated, the pressure must be maintained in the fuel supply manifold to the combustor to avoid two-phase flow. Such flow would produce an uneven fuel flow distribution to the combustor.

Test Condition

The combustor was operated at a simulated supersonic cruise condition of 4 atmospheres pressure, an inlet-air temperature of 838 K (1050° F), and a combustor reference Mach number of approximately 0.075. The desired combustor exhaust-gas temperature was 1478 K (2200° F). Some data were taken at lower fuel flows (lower exhaust-gas temperatures) to obtain higher fuel temperatures. This action was caused by the limitations of the fuel heating system previously described and the fact that the lower heating value per unit weight of Jet-A fuel compared to natural gas requires more fuel flow to obtain the same combustor exhaust-gas temperature.

RESULTS AND DISCUSSION

Data taken during the test program are presented in table II. The NOX emissions data were adjusted to zero inlet-air humidity by multiplying the measured values by the empirical factor e19H, where H is the absolute humidity (g of water/g of dry air), reference 7. To reduce the scatter in the data, the NOX emissions data were also adjusted to the nominal reference Mach number by assuming that NOX varies inversely with Mach number, reference 7.

Effect of Fuel Temperature on Oxides of Nitrogen

The NOX emission index increased with increasing fuel temperature, as shown in figure 6. The increase in NOX with fuel temperature is attributed to increased flame temperature in the primary combustion zone, which is caused by the increased enthalpy of the fuel. Increasing flame temperature increases the rate of formation of nitric oxide (NO) with time, reference 7;
\[ [\text{NO}] = 9.5 k_2 \left\{ \exp \left( \frac{-75.5 \text{kcal/g-mole}}{RT_f} \right) \times [O] \times [N_2] \right\} \]  

(2)

where

\[ [\dot{\text{NO}}] \text{ rate of formation of NO, } \frac{d[\text{NO}]}{dt} \]

\[ k_2 \text{ (13+4) \times 10^{12} \text{ cm}^3 \text{mole}^{-1} \text{sec}^{-1}} \]

\[ R \text{ 1.987 cal/(g-mole)(K)} \]

\[ [\text{NO}] \text{ concentration of NO} \]

\[ [O] \text{ concentration of O} \]

\[ [N_2] \text{ concentration of N}_2 \]

\[ T_f \text{ flame temperature, K} \]

If the concentrations of \( N_2 \) and \( O \) remain constant, the rate equation (2) implies an exponential increase in NO formation with increasing flame temperature.

The rate of increase in \( NO_x \) which might be expected from an increase in fuel temperature can be calculated based on a simplified combustion model. The model specifies that all the \( NO_x \) is formed in a primary combustion zone where the fuel-air ratio is stoichiometric and that all the additional enthalpy of the fuel raises only the primary-zone flame temperature, affecting the rate of formation of NO (eq. (2)). The change in flame temperature may be calculated from the change in enthalpy of the heated fuel:

\[ \Delta T_{\text{flame}} = \frac{\Delta H_{\text{fuel}}}{C_p} \left( \frac{f/a}{1 + f/a} \right) \]  

(3)

where

\[ \Delta T_{\text{flame}} \text{ change in flame temperature} \]

\[ \Delta H \text{ change in enthalpy of the fuel due to heating} \]

\[ C_p \text{ specific heat at constant pressure of the combustion gases at a stoichiometric fuel-air ratio (ref. 8) of about 0.6} \]

\[ f/a \text{ stoichiometric fuel-air ratio} \]

By using equations (2) and (3), the increase in formation rate of NO concentration may be calculated:
\[
\frac{[\dot{\text{NO}}]}{[\text{NO}]}_0 = \frac{\exp \left( \frac{-75.5}{R(T_o + \Delta T)} \right)}{\exp \left( \frac{-75.5}{RT_o} \right)}
\]  

where \([\dot{\text{NO}}]_0\) corresponds to the rate of formation of \([\text{NO}]\) at ambient fuel temperature for a stoichiometric flame temperature \(T_o\).

This calculation was carried out for the inlet-air temperature tested. Figure 7 shows the theoretical result and the data normalized to the value with ambient fuel temperature, 300 K \((80^\circ F)\). The data agree well with the model. The \(\text{NO}_x\) emission index increased approximately 6 percent per 100 K \((180^\circ F)\) increase in fuel temperature.

Comparison with Natural Gas

The rate of increase in \(\text{NO}_x\) fuel temperature is similar to the results obtained with natural-gas fuel in reference 1. Figure 8 shows a comparison of the results obtained with the two fuels. The curves are normalized to the respective \(\text{NO}_x\) emission values at ambient fuel temperature \((300 \text{ K } (80^\circ F))\). The increase in \(\text{NO}_x\) with Jet-A fuel is slightly higher than the increase in \(\text{NO}_x\) with natural-gas fuel. This would be expected from the increased enthalpy available with Jet-A at the same temperature. In a practical aircraft application, however, the increased heat-sink capability of Jet-A could mean lower fuel temperatures delivered to the combustor for a given heat addition to the fuel.

Effect of Overall Fuel-Air Ratio on Oxides of Nitrogen

Figure 9 shows that there was essentially no effect of overall fuel-air ratio on \(\text{NO}_x\) emission index with the fuel at ambient temperature at this test condition. Unpublished data of \(\text{NO}_x\) emissions at other test conditions also indicates that, for this combustor, fuel-air ratio has very little effect on \(\text{NO}_x\) emission index.

Sample Validity

A calculation of the gas sample fuel-air ratio was made for each data point. The ratio of the gas sample fuel-air ratio to the metered fuel-air ratio (fuel-air-ratio ratio) is presented in table II. The maximum data scatter is ±5 percent about a mean of 1.10.
The fact that the mean value is 10 percent high is probably symptomatic of the location of the sampling probes and does not invalidate the trends in the data.

Combustion Efficiency

Combustion efficiency was primarily determined by exhaust-gas sampling and was over 99.96 percent at all test conditions (see table II). A check on the combustion efficiency could be made by measuring the average exhaust-gas temperature with a rotating thermocouple rake. These measurements were made at only limited data points. The enthalpy addition due to heating the fuel was taken into account in the efficiency calculation as determined by exhaust-gas temperature. The thermocouple data supported the results obtained with gas sampling.

Nitric Oxide Content in Oxides of Nitrogen

Nitric oxide (NO) made up approximately 96 percent of the NO\textsubscript{X} emission index at the nominal test condition of 4 atmospheres pressure. Previous tests with this combustor at 6 atmospheres pressure and the same inlet and temperature indicated that the NO made up approximately 91 percent of the NO\textsubscript{X} emission index at those conditions.

SUMMARY OF RESULTS

A test was conducted to determine the effect of increased fuel temperature on the formation of oxides of nitrogen (NO\textsubscript{X}). An annular gas turbine combustor was tested with heated ASTM Jet-A fuel at fuel temperatures from ambient to 700 K (800\textdegree F). Combustor inlet-air conditions simulated conditions at supersonic cruise. Combustor pressure was 4 atmospheres, inlet-air temperature was 838 K (1050\textdegree F), and reference Mach number was 0.075. The NO\textsubscript{X} emission index increased at a rate of 6 percent per 100 K (180\textdegree F) increase in fuel temperature. The result agrees with the results obtained in similar tests run on the same combustor with heated natural gas. The result is also in agreement with the result expected from a simplified combustion model which relates
the increase in the formation rate of nitric oxide to the increase in stoichiometric flame temperature.

Lewis Research Center,
National Aeronautics and Space Administration,
Cleveland, Ohio, October 1, 1973,

REFERENCES


### TABLE I - COMPARISON OF ENTHALPY PROPERTIES OF JET A AND NATURAL GAS

<table>
<thead>
<tr>
<th>Fuel</th>
<th>Enthalpy at two fuel temperatures, J/g of fuel</th>
<th>Lower heating Stoichiometric Fuel-air ratio, J/g</th>
<th>J/G of fuel, Btu/lb</th>
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<th>Pressure, atm</th>
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### TABLE II - EXPERIMENTAL DATA

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<th>Airflow, lb/sec</th>
<th>Inlet total temperature, K</th>
<th>Inlet state, pressure, atm</th>
<th>Emission indices</th>
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aRatio of gas sample to enriched fuel-air ratios
Combustion air available:
289 to 311 K (60° to 100° F);
up to 136 kg/sec (300 lb/sec);
up to 114 N/cm² (165 psia)

Fuel available:
300 K (80° F);
0 to 690 N/cm²
(0 to 1000 psia)

Figure 1. - Test facility.

Figure 2. - Cross section of combustor. Dimensions are in cm (in.).
Figure 3. - Annular ram-induction combustor.
Figure 3. - Closeup view from downstream end.

Figure 4. - Gas sampling probe.
(a) Instrument console.

(b) Schematic diagram.
Figure 5. Exhaust gas analyses system.
Figure 6. - Effect of fuel temperature on NO\textsubscript{X} emission index. Pressure, 4 atmospheres; reference Mach number, 0.075; inlet-air temperature, 838 K; zero humidity.

Figure 7. - Effect of fuel temperature on normalized NO\textsubscript{X} emission index - comparing data with theory. Theoretical flame temperature, approximately 2500 K; specific heat of theoretical flame, approximately 0.60.
Figure 8. Effect of fuel temperature on normalized NO\textsubscript{x} emission index for ASTM Jet-A and natural-gas fuels. Inlet-air temperature, 838 K.

Figure 9. Effect of fuel-air ratio on NO\textsubscript{x} emission index at ambient fuel temperature. Pressure, 4 atmospheres; reference Mach number, 0.075; inlet-air temperature, 838 K; zero humidity.
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—National Aeronautics and Space Act of 1958

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