DEVELOPMENT OF SOLUTION TECHNIQUES FOR
NONLINEAR STRUCTURAL ANALYSIS

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Prepared by

BOEING AEROSPACE COMPANY
RESEARCH AND ENGINEERING DIVISION
SEATTLE, WASHINGTON  98124

R. G. Vos - Technical Leader
J. S. Andrews - Program Leader

Prepared for

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION
GEORGE C. MARSHALL SPACE FLIGHT CENTER
MARSHALL SPACE FLIGHT CENTER, ALABAMA  35812
Nonlinear structural solution methods in the current research literature are classified according to order of the solution scheme, and it is shown that the analytical tools for these methods are uniformly derivable by perturbation techniques. A new perturbation formulation is developed for treating an arbitrary nonlinear material, in terms of a finite-difference generated stress-strain expansion. Nonlinear geometric effects are included in an explicit manner by appropriate definition of an applicable strain tensor. A new finite-element pilot computer program PANES (Program for Analysis of Nonlinear Equilibrium and Stability) is presented for treatment of problems involving material and geometric nonlinearities, as well as certain forms on nonconservative loading.
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1.0 INTRODUCTION

Purpose and Scope of the Study - The present research was undertaken to develop improved techniques for solution of structures with material and geometric nonlinearities, including the limit point and bifurcation behavior which occurs in buckling and collapse problems. Because the effectiveness of such solution techniques has been found to depend strongly on the method used for generating the nonlinear equations, e.g., creation of the system Jacobian matrix, improved equation generation techniques were also emphasized. Available nonlinear analysis methods were evaluated for their current capabilities and their projected long term potentials, and the methods judged to be most promising formed a starting point for development of the techniques presented in this report. Corresponding FORTRAN subroutines were developed and incorporated into the pilot computer program PANES (acronym of the Program for Analysis of Nonlinear Equilibrium and Stability) for checkout and evaluation. The equation generation and solution techniques are within the framework of the finite element structural discretization method.

1.1 General Philosophy and Evaluation of Methods

Criteria - Structural solution methods available in the current literature were initially evaluated for this study based on four general criteria:

1. A high degree of automation which minimizes the burden on the user.
2. Cost effectiveness for large size problems.
3. The use of an effective incremental technique which allows the user to follow and plot the structural response path.
4. Achievement of accuracy by a self-correcting characteristic, which assures that the true solution is approached at each point where results are desired.
During the study it was decided that recent advancements in structural theory made it timely to broaden the applicability of the developed equation generation and solution techniques by including a fifth requirement:

5. An efficient treatment of large-strain problems, and of arbitrary nonlinear elastic or inelastic materials.

Classification of Methods - The current literature contains a very broad variety of nonlinear solution methods, and even the specialized requirements for nonlinear structural solutions have not resulted in consensus on a best method or methods. On the other hand, certain types of highly nonlinear problems are presently receiving considerable attention (for example, stability analyses and large-strain effects with arbitrary nonlinear materials), and such problems tend to eliminate certain methods from consideration while giving some direction to future research and development.

Most of the nonlinear structural solution methods can be broadly grouped into three classes:

1. Methods which use only the initial (constant) stiffness of the structure, and rely on iteration with the calculation of residual (unbalanced) forces to achieve the correct solution. The loading may be applied incrementally or in a single step.

2. Methods which form the Jacobian (tangent stiffness) matrix at a series of load increments, without iteration; also, various combined incremental and iterative algorithms which update the Jacobian at each step or periodically.

3. Higher-order methods (perturbation approaches or various numerical integration schemes) which employ higher-order derivative relations in addition to the first-order Jacobian coefficients.
In many respects, the above ordering of classes is according to increasing sophistication and greater capability. For example, the class 3 methods are especially suited for analysis of complicated limit point and postbuckling problems. As might be expected, the historical development of nonlinear solution methods has shown some tendency to progress from original class 1 techniques to those of classes 2 and 3.

1.2 Previous Developments and Present Work

Historical Development - Early finite-element work in nonlinear structural analysis began with a paper by Turner, Dill, Martin and Melosh (1960). This work incorporated nonlinear geometric effects within the so-called "geometric" stiffness matrix, and various incremental and iterative solution procedures were recognized by the authors. In the initial attempts at nonlinear solutions which followed, there was a natural tendency to generalize existing linear capabilities. This usually led to iterative approaches and use of the initial constant stiffness matrix, with calculation of the nonlinearities as additional load terms. The geometric stiffness matrix, however, formed the basis for eigenvalue buckling analyses. A more consistent and theoretical basis for the geometric stiffness matrix was investigated by Gallagher and Padlog (1963), who used a strain-energy derivation with the same displacement functions for both the linear and nonlinear stiffness terms. Several formulations for nonlinear beam and plate analysis soon followed. Mallett and Marcal (1968) presented a unifying basis for formulating large displacement problems, by deriving the total strain energy as a function of nodal displacements and including previously neglected nonlinear terms. Meanwhile, developments were proceeding in the area of material nonlinearities, for example the plasticity work of Argyris (1965) and Marcal (1968), and nonlinear elastic analysis by Oden and Kubitza (1967).
A consensus on solution methods did not appear, however, and different approaches were emphasized by different research groups. Zienkiewicz and coworkers popularized some of the class 1 solution methods, and Zienkiewicz et al. (1969) presented a particular method called the "initial stress" residual-load method. This work was followed more recently by techniques for improving the iterative convergence of such methods, e.g. Nayak and Zienkiewicz (1972). However, the paper by Zienkiewicz and Nayak (1971) presents a quite general formulation for various class 2 methods with application to combined geometric and material large-strain nonlinearities. Considerable work in geometric and material nonlinearity has been done at Brown University, for example Marcal (1969), and McNamara and Marcal (1971). Researchers there have tended to favor class 2 methods without iteration, although the use of one or two iterations in each load step has been suggested as a way of increasing accuracy. Combined incremental and iterative class 2 techniques (Newton Raphson iteration, for example) have been employed for analysis of highly nonlinear material and geometric nonlinearities, including some stability problems, by Oden and Key (1970), Sandidge (1973), and Key (1974). It would certainly appear that for such problems the class 1 methods at least are highly unsuitable. A number of nonlinear survey and development papers have been written at Texas A & M University, including Haisler et al. (1971), Stricklin et al. (1972), Stricklin et al. (1973) and Tillerson et al. (1973). These papers provide a detailed investigation of various class 1 and 2 methods, as well as certain class 3 methods which rely on numerical integration schemes. Although perturbation procedures are not tested by these authors, it is suggested in one of the early papers that perturbation techniques would be very time consuming for cases with large degrees of freedom, while a more recent paper notes that these techniques require further evaluation before they will be accepted by structural analysts. Many other researchers working in the areas of limit point and bifurcation stability problems, however, have concentrated on perturbation methods, reviving the
original theoretical developments in that area by Koiter (1945). Haftka et al. (1970) use an extension of Koiter's perturbation theory in a solution approach called the "modified structure method." Morin (1970) uses perturbation techniques in developing higher-order predictor and corrector algorithms for analysis of geometrically nonlinear shells. Gallagher and Mau (1972) and Mau and Gallagher (1972) establish procedures for limit point and postbuckling analysis based on perturbation expansions and the evaluation of determinants, which employ a combination of class 1, 2 and 3 solution techniques. A number of other perturbation developments of a more theoretical nature are included in the references and bibliography section of this report.

Much of the present diversity in nonlinear solution methods can be attributed to a desire to further investigate the potentials of all methods and to compare the results obtained from them. However, the comparisons and evaluations which are presented often disagree in their conclusions as to the effectiveness of a particular method. It must be surmised that the evaluation of nonlinear solution methods is necessarily influenced by the previous experiences and preferences of the researcher, by the degree of sophistication in his various solution method tools, and by the type of problems toward which his interests are directed.

Direction of Present Work - Because the present work was directed toward obtaining techniques whose applicability included the more highly nonlinear structures, a decision was made to eliminate from consideration the constant-stiffness methods of class 1. Although schemes have been proposed for extending these methods to more severe nonlinearities, it must be said that the arguments given are not convincing. In fact, when the structural system has advanced into a highly nonlinear state, the initial constant portion of the stiffness does not really possess any more significance than that provided by an arbitrary positive definite matrix; it can not be expected that a technique based on this
matrix will be of any significant value in advancing the solution beyond the current state. It was also decided in the present work to reject those methods of a non self-correcting nature, i.e., methods which do not involve an iterative calculation of the unbalanced or "residual" forces, which gives an indication of accuracy and allows the solution to be improved. Although such methods are sometimes effective, they can lead to serious errors in the computed results, especially for path dependent problems. A third group of methods eliminated from consideration were those which use the solution data generated at several previous solution points. Such methods essentially extrapolate the previous data, either by some numerical integration formula or by a curve fitting approach. These methods require storage of previous data and are usually not self-starting. However the main objection to their use would seem to be that the same type of capability is provided by perturbation methods, which more accurately evaluate the path direction and are more generally applicable to a wide range of highly nonlinear problem types (e.g., those involving path discontinuities such as bifurcation points).

With these considerations, the methods which remain for development include methods of "incremental loading", Newton Raphson iteration and its modifications involving only periodic updating of the Jacobian, and higher-order methods including various orders of predictor and corrector algorithms. In order to make the current methods applicable to cases of large strain and arbitrary nonlinear materials, the equation generation process is accomplished in the present work by a finite difference expansion procedure. It is found that generation of the nonlinear equations by this means within a perturbation context provides a unifying basis for definition of the nonlinear solution terms, including as special cases the first-order Newton Raphson and incremental loading methods, as well as almost an unlimited variety of higher-order solution techniques. The perturbation
procedures have the advantage of a sound theoretical basis in classical developments, and lend themselves readily to both limit point and postbuckling problems as well as to simple nonlinear behavior without critical points.
2.0 DETERMINATION OF THE EQUILIBRIUM PATH: A GENERALIZATION OF STATIC PERTURBATION TECHNIQUES

In this Section the theory and techniques are developed for following the nonlinear equilibrium path of a structure under prescribed loading. It is assumed that the equilibrium path is continuous and unique, although limit point behavior is allowed (the non-uniqueness due to bifurcation of the equilibrium path is considered in Section 3. The development follows the "static perturbation method" which was recognized and established in concrete form by Sewell (1965). The present work generalizes previous structural solution techniques based on the method to allow effective treatment of arbitrary nonlinear materials. The resulting formulation is shown to provide a quite general and unifying basis for solution of nonlinear structures, including geometric and material nonlinearities as well as certain forms of nonconservative loading. A summary of the formulation is contained in the paper by Vos (1974).

2.1 Description of Nonlinearities

An important characteristic of the present method is a preliminary separation of the nonlinear material and geometric effects, which minimizes the required number of perturbation expansion terms, and also increases numerical accuracy.

Material Effects - The nonlinear material effects are described by expanding the stress about a known equilibrium configuration:

\[ \sigma_i = \sigma_i^* + D_{ij}^* \Delta \varepsilon_j + D_{ijk}^* \Delta \varepsilon_j \Delta \varepsilon_k + \ldots \]  

which provides the stress, \( \sigma \), in terms of the incremental strain, \( \Delta \varepsilon \). Here and throughout this work, an asterisk (*) denotes quantities evaluated at a known equilibrium state, and \( \Delta \) denotes an incremental quantity. In (2-1) \( \sigma^* \) is the initial stress, while
DO* and DL* are 2nd and 3rd order incremental stress-strain tensors, respectively. This type of expansion can be developed numerically for a general nonlinear elastic or inelastic material, by an efficient finite difference or Taylor series evaluation. Complete symmetry of the D tensors can be used to advantage if they are derivable from a strain-energy function, or in certain other cases such as that of associative plasticity. These considerations are discussed by Zienkiewicz and Nayak (1971) in a development which employs only the 2nd order (DO) tensor. In any case, the tensors can be made symmetric in the j, k and any higher order indices.

Geometric Effects - The nonlinear geometric effects are included through a definition of finite element displacement functions and an appropriate strain tensor, giving

\[ \theta_i = G_{ij} q_j \]  \hspace{2cm} (2-2a)

\[ \epsilon_i = A_{0ij} \theta_j + \frac{1}{2} A_{ijk} \theta_j \theta_k \]  \hspace{2cm} (2-2b)

Here \( q \) are the element generalized (nodal) displacements of an element, \( G \) is obtained by differentiating the assumed displacement functions, \( \theta \) are the displacement derivatives at any point, while \( A_0 \) and \( A_l \) are constant coefficients which define the strain tensor with \( A_{ijk} = A_{ikj} \). The explicit form (2-2b) of the strain tensor will be seen to simplify later manipulations.

† The best approach developed thus far is a forward difference scheme, which requires a minimum number of function evaluations and allows an arbitrary difference size for each independent variable. Explicit coefficients have been derived for such expansions of either linear, quadratic or cubic form, and in terms of an arbitrary number of variables. Details are provided in Appendix A. It may be noted that for certain problems involving incompressible materials, the hydrostatic stress must be included as an independent parameter in addition to the strains.
Advantages of Present Approach - The present approach defines all nonlinearities through the form of (2-1) and (2-2), rather than through a direct expansion of the nodal displacements such as that used in the investigation of Oden and Key (1970). The present approach appears to offer substantial advantages, because it allows implementation of perturbation theories into limit point and bifurcation analysis, without involving a huge number of terms and formidable algebraic operations. As a practical matter, it should also be noted that a numerical expansion based on displacements often causes severe problems with accuracy of the expansion coefficients, due to large differences in magnitude between individual displacement limits (e.g., between the membrane and bending freedoms of a plate or shell), and the selection of accurate finite difference sizes then becomes difficult. Accuracy is more easily obtained in an expansion of the type (2-1), because the strain limits tend to be of the same order of magnitude.

2.2 Formulation of Equilibrium Equations

Virtual Work Statement - The principle of virtual work, which is valid for arbitrary nonlinear materials and nonconservative systems, is employed to obtain equilibrium equations for the system of finite elements. The formulation is developed here for a conservative system, and nonconservative effects are treated in Appendix B. The equivalence of external and internal virtual work, relates the generalized nodal forces \( p \) and displacements \( q \) of a particular finite element, in the element equilibrium equation

\[
\delta q_i p_i = \int_V \delta \varepsilon_a \sigma_a \, dV \tag{2-3}
\]

which holds along any equilibrium path in the neighborhood of the reference equilibrium (*) configuration. Here \( \delta \varepsilon \) and \( \delta q \) are kinematically consistent variations, and from equation (2-2)
\[ \delta \varepsilon_i = G_{ij} \delta q_j \quad (2-4a) \]
\[ \delta \varepsilon_i = (A_{0ij} + A_{1ij} \theta_k) \delta \theta_j \equiv A_{ij} \delta \theta_j \equiv B_{ij} \delta q_j \quad (2-4b) \]

The integral in equation (2-3) is taken over the volume of the element, and it is to be noted that a proper definition of stress and strain is required to give the correct evaluation of internal work. One approach for accomplishing this is a formulation of Lagrangian strain and second Piola-Kirchoff stress integrated over the undeformed volume, e.g. see Oden and Key (1970).

**Basic Equilibrium Equation** - Substituting for \( \delta \varepsilon \), and noting that (2-3) must be satisfied for arbitrary variations \( \delta q \), provides the basic equilibrium equation for the element, as

\[ p_i = \int_B \sigma_{ai} \, dV = \int_B G_{mi}^{ai} (A_{0am} + A_{lamn} \theta_n) \sigma_a \, dV \quad (2-5a) \]

In order to merge the element equations into the system equations, the usual type of finite element transformation is applied. The system forces and displacements will be denoted by the capitals \( P \) and \( Q \), respectively, and the system basic equilibrium equation corresponding to (2-5a) is written as

\[ p_i = \int_B B_{ai} \sigma_a \, dV = \int_B G_{mi}^{ai} (A_{0am} + A_{lamn} \theta_n) \sigma_a \, dV \quad (2-5b) \]

where now it must be understood that the integral is summed over all elements while applying the proper element-system nodal transformations. With this understanding, the element and system quantities will here be used interchangeably.

**Derivative Relations** - Equations (2-5) may now be differentiated as many times as desired with respect to some suitable path parameter. Toward that end, it is useful to record here the following typical derivative relations, where an overdot ('') denotes differentiation with respect to the path parameter.
\[ \dot{\sigma}_{ab} = D_{ab} \epsilon_{b} + 2D_{ab} \epsilon_{b} \Delta \epsilon + \ldots \]

\[ \ddot{\sigma}_{ab} = D_{ab} \ddot{\epsilon}_{b} + 2D_{ab} \epsilon_{b} \Delta \epsilon + 2D_{abc} \epsilon_{b} \Delta \epsilon + \ldots \]

\[ \dot{\epsilon}_{a} = B_{ai} \dot{q}_{i} \]

\[ \ddot{\epsilon}_{a} = B_{ai} \ddot{q}_{i} + B_{ai} \dot{q}_{i} = B_{ai} \ddot{q}_{i} + A_{amn} \theta \theta \]

and at the reference configuration \((\Delta \epsilon = 0)\), we have

\[ \dot{\sigma}_{ab} = D_{ab} \epsilon_{b} \]

\[ \ddot{\sigma}_{ab} = D_{ab} \ddot{\epsilon}_{b} + 2D_{abc} \epsilon_{b} \Delta \epsilon \]

\textbf{First Order Equilibrium Equation} - Differentiating equation (2-5b) once, and evaluating at the reference equilibrium (*) configuration, gives

\[ \dot{p}_{i} = \int (B_{ai} \dot{\sigma}_{a} + B_{ai} \dot{\sigma}^{*}) \, dV \]  

Substituting from relations (2-6b) gives

\[ \dot{p}_{i} = \int (G_{mi} \dot{\epsilon}_{a} A_{amn} G_{nj} \dot{q}_{j} + B_{ai} D_{ab} \dot{\epsilon}_{b} \dot{q}_{j}) \, dV \]  

This is the first order equilibrium equation, which may be written in the form

\[ \dot{p}_{i} = K_{ij} \dot{q}_{j} \]  

where

\[ K_{ij} = \int (G_{mi} \epsilon_{a} A_{amn} G_{nj} + B_{ai} D_{ab} \epsilon_{b}) \, dV \]

and

\[ B_{ai} = G_{mi} A_{amn} = G_{mi} (A_{0 am} + A_{amn} \theta^{*}) \]
The "tangent stiffness" relation (2-7c) is equivalent to the incremental matrix formulation of Zienkiewicz and Nayak (1971), although the tensor form given here shows perhaps more clearly the symmetry and differentiability properties of the tangent (Jacobian) matrix \( K_0^* \). The first contribution to \( K_0^* \) is due to the initial stresses during changing geometry, and is always symmetric in form. The second contribution is due to the incremental stress-strain relation, and its symmetry depends on symmetry of the matrix \( D_0^* \).

**Second Order Equilibrium Equation** - A second differentiation of (2-5b) and evaluation at the reference configuration, gives

\[
\dddot{P}_i^* = \int \left( \dot{B}^* \sigma^* + 2 \dot{B}^* \dot{\sigma}^* + B^* \dddot{\sigma}^* \right) dV \quad (2-8a)
\]

Substituting from relations (2-6b) gives

\[
\dddot{P}_i^* = \int \left\{ \frac{G_m}{a} \sigma^* \dot{A}_{mn} G_{nj} \dddot{\sigma}^* + 2 \frac{G_m}{a} \dot{A}_{mn} \dddot{\theta}^* D_0^* \dddot{\epsilon}^* \right. \\
+ \left. B^* \left( D_0^* B^* \dddot{\theta}^* + D_0^* \dot{A}_{brs} \dddot{\epsilon}^* + 2 D_1^* \dddot{\epsilon}^* \right) \right\} dV \quad (2-8b)
\]

This is the second order equilibrium equation, which may be written in the form

\[
\dddot{P}_i^* = K_0^* \dddot{Q}_i^* + P_1^* \quad (2-8c)
\]

where \( P_1^* \) is a pseudo force term given by

\[
P_1^* = \int G_m \left( D_0^* \left( A^* \dot{A}_{brs} \dddot{\epsilon}^* + 2 \dot{A}_{mn} \dddot{\theta}^* \dddot{\theta}^* \right) + 2 D_1^* \dddot{\epsilon}^* \right) dV
\]
2.3 Solution of Equilibrium Equations

Incremental Load and Path Parameters - An increment of conservative loading is defined by

\[ \Delta P_i = \Lambda P_i^0, \quad P_i = \Delta P_i^0, \quad \text{etc.} \quad (2-9) \]

using a variable load parameter \( \Lambda \) and constant nodal load distribution \( P_i^0 \).

Taylor series expansions are then used to approximate both the incremental load parameter \( \Lambda \) and displacements \( \Delta Q \):

\[ \Lambda = \dot{\Lambda} S + \frac{1}{2} \ddot{\Lambda} S^2 + \ldots \quad (2-10a) \]
\[ \Delta Q_i = \dot{Q}_i S + \frac{1}{2} \ddot{Q}_i S^2 + \ldots \quad (2-10b) \]

In order to handle limit point situations within the present formulation, the path parameter \( S \) is here taken as defined by

\[ S^2 = i K^{ij}_i \Delta Q_i \Delta Q_j \geq 0, \quad i = \pm 1 \quad (2-11) \]

with the requirement that \( K^{ij}_i \), evaluated at the beginning of each load increment, be nonsingular (either positive or negative definite). Without any loss of generality, additional requirements imposed at every path point, \( S \), are that

\[ \dot{S} = \ddot{S} = 1 \]
\[ \dddot{S} = \dddot{S} = \ldots = 0 \]

Successive differentiation of (2-11) provides the relations

\[ 2S \dot{S} = i K^{ij}_i (\dot{Q}_i \Delta Q_j + \Delta Q_i \dot{Q}_j) \]
\[ 2 \dddot{S}^2 = i K^{ij}_i (\ddot{Q}_i \Delta Q_j + 2 \dot{Q}_i \dot{Q}_j + \Delta Q_i \dddot{Q}_j) \quad (2-12a) \]
\[ 0 = i \mathbf{K}_j^* \left( \dddot{Q}_i \Delta Q_i + 3 \dddot{Q}_i \dddot{Q}_j + 3 \dot{Q}_i \dddot{Q}_j + \Delta Q_i \dddot{Q}_j \right) \]  

and evaluation at the reference state \( (S = \Delta Q_i = 0) \), yields

\[ \dddot{Q}^2 = 1 = i \mathbf{K}_j^* \dot{Q}_i^* \dot{Q}_j^* \]

\[ 0 = i \mathbf{K}_j^* \left( \dddot{Q}_i^* \dot{Q}_j^* + \dddot{Q}_j^* \dot{Q}_i^* \right) \]  

\[ (2-12b) \]

It may be noted that relations (2-12) hold for the general case of an unsymmetric \( \mathbf{K}_j^* \) matrix.

**Determination of Rate Quantities** - In order to implement various solution techniques, the equilibrium equations (2-7c) and (2-8c) must be used to determine the load and displacement rates.

Multiplying (2-7c) by \( \dddot{Q}^* \), and making use of (2-9) and (2-12b), gives

\[ i = \dot{\mathbf{\Lambda}}^* \dot{P}_i^* \]  

\[ (2-13a) \]

Solving (2-7c) for \( \dot{Q}^* \) gives

\[ \mathbf{K}_j^* \dot{Q}_i^* = \dot{Q}_i^* = \dot{\mathbf{\Lambda}}^* \dot{Q}_i^* \]  

\[ (2-13b) \]

and substituting \( \dot{\mathbf{\Lambda}}^* \dot{Q}_i^* \) for \( \dot{Q}^* \) in (2-13a) gives

\[ \dot{\mathbf{\Lambda}}^* = i / (Q_i^* P_i^*) \]  

\[ (2-13c) \]

where now \( i \) is chosen to make \( \dot{\mathbf{\Lambda}}^* \) positive. Multiplying (2-8c) by \( \dddot{Q}^* \), and again making use of (2-9) and (2-12b), gives

\[ \dddot{Q}_i^* \dddot{P}_i^* = \mathbf{K}_j^* \dddot{Q}_i^* \dddot{Q}_j^* + \dddot{Q}_i^* \dot{P}_l^* = -\mathbf{K}_j^* \dddot{Q}_i^* \dddot{Q}_j^* + \dddot{Q}_i^* \dot{P}_l^* \]  

\[ (2-13a) \]
Solving (2-8c) for $\ddot{Q}^*$ gives

$$\ddot{Q}^*_i = K_{i}O^{-1}_* \left( \ddagger^* P^{i}_j - P^{i}_j \right) \equiv \ddagger^* Q^*_i - Q^*_i$$

(2-14b)

Substituting (2-14b) into (2-14a) with the use of (2-13c) then provides the result

$$\ddagger^* = i\ddagger^* \frac{(P^{i}_1 Q^*_1 + P^{i}_1 Q^*_1) / 2}{2}$$

(2-14c)

after which $\ddot{Q}^*$ is obtained directly from (2-14b).

It is to be noted that a solution for the rates $\dot{A}^*$, $\ddagger^*$, $\dot{Q}^*$ and $\ddot{Q}^*$ (and higher order rates if desired by similar calculations) requires only a single formation and decomposition of the matrix $K_0^*$.

Solution Procedures - Once the load and displacement rates have been determined to a desired order, many different solution procedures can be applied in tracing the nonlinear equilibrium path of the structure. The first order rates allow solution by methods of incremental loading (with or without evaluation of residual forces and corrective iterations), and Newton Raphson iteration where the Jacobian is re-evaluated at each iteration. Various combinations of incrementation and iteration, with periodic updating of the Jacobian are of course possible. The second order rates allow the use of a second order predictor. The additional cost of the 2nd order predictor is associated with the Pl psuedo-force term, whose evaluation is performed at the elemental level with a cost roughly proportional to that of a single "residual force" evaluation. The cost of evaluating Pl by the form of (2-8c) is only linearly proportional to the number of integration points within an element, so that this technique is effective even for elements having complex geometry and large degrees of freedom.
Such a predictor has been found to be very useful during the present study, and although the PANES computer program allows use of various predictor-corrector options, the second order predictor almost always appears to be more efficient for cases of substantial nonlinearity. Since the second order rates are valid at any reference equilibrium configuration, they may be applied in a corrector technique, at a state where the system is in "equilibrium" under the applied loads plus a set of unbalanced residual loads. Thus convergence could be considerably accelerated if the second order relations were computed and used at each iteration, although the cost per iteration would also increase considerably. Higher order predictor-corrector relations are obviously possible as well, and the best type of solution capability would probably be a program in which more or less arbitrary options are allowed for the order of predictor and corrector, the frequency with which the Jacobian is updated, and the number of iterations to be performed per update. Although these considerations will not be discussed in any more detail here, the PANES program is at least a step in that direction, and makes available various options using the first and second order rate relations.

Limit Point and Step Size Considerations - A major advantage of a 2nd order predictor is that, with little increase in computational effort, it provides greatly increased prediction accuracy and allows larger load steps to be taken. In addition, it enables the traversing of limit points and provides various techniques for automatic selection of the load step size.

In the vicinity of a limit point, the load rate relation

\[ \dot{\lambda} = \dot{\lambda}^* + \dot{\lambda}^*S \]  

(2-15a)
is used. Also from (2-10a) the path value for given \( \Lambda \) is

\[
S = \{ \frac{-\Lambda^* + (\Lambda^* + 2\Lambda^*)^{1/2}}{\Lambda^*} \}/\Lambda^* \geq 0 \tag{2-15b}
\]

At a limit point \( \Lambda = 0 \), so that from (2-15a) the critical path value is

\[
S^C = -\Lambda^*/\Lambda^* \tag{2-15c}
\]

Using these relations the limit point can be traversed when \( \Lambda \) is within some specified fraction of its critical value.

With regard to automatic selection of a general load step size, the following predictor relationships are noted.

\[
\Delta \Lambda = \Lambda S + \frac{1}{2} \Lambda S^2 \tag{2-16a}
\]

\[
\Delta Q_i = Q_i S + \frac{1}{2} Q_i S^2 \tag{2-16b}
\]

Here the quadratic terms give an indication of the accuracy of the linear predictor, but because of the truncation of higher order terms there is no indication of accuracy for the quadratic predictor. The rationale used in the PANES program implementation is therefore to select a load step size which limits the quadratic contributions to some specified factor times the linear contributions. Specifically, change in slope of the load parameter during a load step is approximated by

\[
\Delta \Lambda = \Lambda S \tag{2-17a}
\]

and the ratio of slope change to average slope is

\[
\Delta \Lambda / \Lambda_{\text{average}} = \Lambda S / (\Lambda + \frac{1}{2} \Lambda S) \tag{2-17b}
\]
This slope ratio is specified as a given allowable magnitude, in order to prevent over-prediction in (2-16a) of the behavior beyond accurate values. A similar step size restriction is employed based on (2-16b) for displacement rates.
3.0 DETERMINATION OF BIFURCATION AND THE POSTBUCKLING PATH

This section considers the identification of bifurcation points in the load-displacement path of a structure, and the prediction of the postbuckling path beyond these points. The formulation follows the approach of Section 2 for representing geometric nonlinearities and an arbitrary nonlinear material. The effects of nonconservative load on bifurcation and postbuckling are treated in Appendix B.

3.1 Description of the Postbuckling Path

We consider behavior of the type shown in Figure 3-1, which is a plot of the incremental load parameter $\Lambda$ for a structure versus its incremental displacements $\Delta Q$, shown here conceptually for a single degree of freedom system. The point 0 represents a reference equilibrium configuration ($Q = Q^*, \Lambda = \Delta Q = 0$). Travel along the "fundamental" and "postbuckling" paths is measured by suitable path parameters $S$ and $R$, respectively. Thus $S$ has a value of zero at point 0, while $R$ takes on a zero value at the critical bifurcation point C.

![Figure 3-1: Fundamental and Postbuckling Paths](image)
We follow the terminology of Mau and Gallagher (1972) and use a "sliding coordinate" system to describe the various fundamental and postbuckling quantities. For a given value of \( \Lambda \), a point on the fundamental path has associated quantities whose values are denoted by \( (\cdot)^{f} \), while additional values at the corresponding point on the postbuckling path are denoted by \( (\cdot)^{P} \). Thus total values on the postbuckling path are denoted by \( (\cdot)^{f} + (\cdot)^{P} \), and we write for the postbuckling path

\[
\begin{align*}
Q &= Q^{f} + Q^{P} \\
\Delta Q &= \Delta Q^{f} + \Delta Q^{P} \\
\Delta \varepsilon &= \Delta \varepsilon^{f} + \varepsilon^{P} \\
\sigma &= \sigma^{f} + \sigma^{P} \\
\text{etc.}
\end{align*}
\]

where the \( \Lambda \) quantities are increments from the fundamental reference configuration.

We will refer to the \( (\cdot)^{f} \) and \( (\cdot)^{P} \) values as the "fundamental" and "postbuckling" values, respectively, and to their sums as the "total" values.

### 3.2 Formulation of Postbuckling Equilibrium Equations

**Basic Equilibrium Equation** - Because the postbuckling path is an equilibrium path, and equation (2-5b) is valid for a point on any equilibrium path, we may write the postbuckling equilibrium equation as

\[
P_{i} = P_{i}^{f} + P_{i}^{P} = \int_{V} B_{ai} \sigma \, dV = \int_{V} (B_{ai}^{f} + B_{ai}^{P}) \left( \sigma_{a}^{f} + \sigma_{a}^{P} \right) \, dV \quad (3-2)
\]
Recognizing that $P_i = P_i$ for a given value of $\lambda$ with conservative loading, and subtracting out terms in equation (3-2) which are zero because they collectively satisfy the fundamental equilibrium equation, provides the desired form of the postbuckling equilibrium equation as

$$p_i^P = 0 = \int \left( f \sigma^P_{ai} + B^P_{ai} (\sigma^f_{ai} + \sigma^P_{ai}) \right) dV$$

(3-3)

**Derivative Relations** - We now record the following typical derivative relations, where a prime (') denotes differentiation with respect to the postbuckling path parameter $R$.

$$\sigma_a^P = \sigma_a - \sigma_a^f = D0^* \varepsilon_a^P + D1^* \varepsilon_a \left( 2\varepsilon^P_{bc} \varepsilon^f_{abc} + \varepsilon^P_{bc} \varepsilon^f_{abc} \right)$$

$$\sigma_a^P = D0^* \varepsilon_a^P + D1^* \varepsilon_a \left( 2\varepsilon^P_{bc} \varepsilon^f_{abc} + \varepsilon^P_{bc} \varepsilon^f_{abc} \right)$$

$$\varepsilon_a^P = \varepsilon_a^f = B_{ai} q_i^f - B_{ai} q_i$$

and at the critical point ($\sigma_a = \sigma_a^f$, $B_{ai} = B_{ai}^f$, $\sigma_a^P = B^P_{ai} = 0$),

3-3
we have

\[
\sigma_\alpha' = D_0 \delta_{\alpha\beta} \epsilon_\beta' + 2D_1 \epsilon_{\alpha\beta} \epsilon_\beta' \Delta \epsilon_f^f = D_0 \epsilon_{\alpha\beta}'
\]

\[
\sigma_\alpha'' = D_0 \delta_{\alpha\beta} \epsilon_\beta'' + D_1 \epsilon_{\alpha\beta} \epsilon_\beta' \Delta \epsilon_f^f + 4 \epsilon_{\alpha\beta} \epsilon_\beta' \epsilon_f^f + 2 \epsilon_{\alpha\beta} \epsilon^f
\]

\[
= D_0 \epsilon_{\alpha\beta}' + D_1 \epsilon_{\alpha\beta} \epsilon_\beta' (4 \epsilon_{\alpha\beta} \epsilon_\beta' \epsilon_f^f + 2 \epsilon_{\alpha\beta} \epsilon^f)
\]

\[
\sigma_\alpha''' = D_0 \delta_{\alpha\beta} \epsilon_\beta''' + D_1 \epsilon_{\alpha\beta} \epsilon_\beta' \Delta \epsilon_f^f + 6 \epsilon_{\alpha\beta} \epsilon_\beta' \epsilon_f^f
\]

\[
+ 6 \epsilon_{\alpha\beta} \epsilon^f
\]

\[
= D_0 \epsilon_{\alpha\beta}' + D_1 \epsilon_{\alpha\beta} \epsilon_\beta' (6 \epsilon_{\alpha\beta} \epsilon_\beta' \epsilon_f^f + 6 \epsilon_{\alpha\beta} \epsilon^f)
\]  

\[
\epsilon_\alpha' = B^f q_i^i = B^* q_i^i + A_l \epsilon_{\alpha\beta} \epsilon_\beta^f
\]

\[
\epsilon_\alpha'' = B^f q_i^i + B^* q_i^i - B^* f q_i^i = B^f q_i^i
\]

\[
+ A_l \epsilon_{\alpha\beta} \epsilon_\beta^f (2 \epsilon_{\alpha\beta} \epsilon_\beta^f + \epsilon_{\alpha\beta} \epsilon^f)
\]

\[
\epsilon_\alpha''' = B^f q_i^i + B^* q_i^i + 2B^* q_i^i - B^* f q_i^i - 2B^* f q_i^i
\]

\[
= B^f q_i^i + A_l \epsilon_{\alpha\beta} \epsilon_\beta^f (3 \epsilon_{\alpha\beta} \epsilon_\beta^f + 3 \epsilon_{\alpha\beta} \epsilon^f)
\]

\[
+ 3 \epsilon_{\alpha\beta} \epsilon^f
\]

**First Order Equilibrium Equation** - Differentiating the basic postbuckling equilibrium equation (3-3), and evaluating at the critical bifurcation point, gives

\[
P^f_i = 0 = \int_V (B^f a_i^a + B^* \phi_i^a) \, dV
\]  

(3-5a)
Substituting for $\sigma'P$ and $B'P$ gives

$$0 = \int_V \left\{ B_{ai}^f (D_{0*}^* \varepsilon_{ab}^* P + 2D_{1*}^* \varepsilon_{abc}^* P \Delta\varepsilon_{bc}^f) + G_{mi} A_{amn} \theta_{P} \sigma_{a}^* \right\} dV \quad (3-5b)$$

Substituting for $B_{ai}^f$, and using the relations which express $\theta^P$ and $\varepsilon^P$ in terms of $q^P$, gives

$$0 = \int_V \left\{ (B_{ai}^* + G_{mi} A_{amn} \Delta\varepsilon_{bc}^f) (D_{0*}^* (B_{ai}^* q^P + A_{bns}^* n^P \Delta\varepsilon_{s}^f) + 2D_{1*}^* \varepsilon_{abc}^* P \Delta\varepsilon_{bc}^f) + G_{mi} A_{amn} (G_{nj} q_{j}^P \sigma_{*}^a + \varepsilon_{P} (D_{0*}^* \Delta\varepsilon_{b}^f + D_{1*}^* \Delta\varepsilon_{b}^f \Delta\varepsilon_{c}^f)) \right\} dV \quad (3-5c)$$

This is the first order postbuckling equilibrium equation, which may be written in the form

$$0 = K_0^*_{ij} Q^P_{j} + P^l_1 \quad (3-5d)$$

where again

$$K_0^*_{ij} = \int_V (G_{mi} A_{amn} G_{nj} + B_{ai}^* D_{0*}^* B_{ai}^*) dV$$

and

$$P^l_1 = \int_V G_{mi} (D_{0*}^* (A_{am}^* A_{am}^* \theta^P \Delta\varepsilon_{bc}^f + A_{am}^* \varepsilon^P \Delta\varepsilon_{bc}^f) + A_{am}^* \Delta\varepsilon_{bc}^f \theta^P) + D_{1*}^* (2A_{am}^* \varepsilon^P \Delta\varepsilon_{bc}^f + 2A_{am}^* \varepsilon^P \Delta\varepsilon_{bc}^f \Delta\varepsilon_{c}^f) + A_{am}^* \Delta\varepsilon_{bc}^f \Delta\varepsilon_{c}^f \theta^P) dV$$

Equation (3-5d) is an eigenequation form, in terms of the unknown critical values $\Delta\theta$ and $\Delta\varepsilon$, which is suitable for solution by power iteration.
Alternatively, the eigenequation may be written in the form

\[ 0 = K_{ij} Q_i^P = (K_{ij}^* + \Delta K_{ij}) Q_i^P \]  

(3-5e)

where \( K \) is the Jacobian at the critical point, and is given by

\[ K_{ij} = \int_V (G_{mi} a_{1} a_{m} a_{n} n_{j} + B_{ai} D_{0} a_{b} b_{nj}) \, dV \]

with the \( a, D_0 \) and \( B \) quantities evaluated at the critical point. (3-5e) may be solved directly for the eigenvector \( Q^P \), provided that the critical values of \( a, D_0 \) and \( B \) have been previously determined (as in the method proposed by Mau and Gallagher (1972)). This equation may also be solved by expressing \( \Delta K \) as a Taylor series expansion in the fundamental path parameter, giving a form suitable for solution by one of the many "direct" eigensolution methods.

**Second Order Equilibrium Equation** - A second differentiation of the postbuckling equilibrium equation (3-3) and evaluation at the critical point, gives

\[ p_{i}''^P = 0 = \int_V \left\{ 2B_{ai}^o \sigma_{a}^o + B_{ai}^o s_{a}^o P + B_{ai}^o P_{a} + 2B_{ai}^o (\sigma_{a}^o + s_{a}^o) \right\} \, dV \]  

(3-6a)

Substituting for \( \sigma''^P \) and \( B''^P \) gives

\[ 0 = \int_V \left\{ 2G_{mi} a_{1} a_{m} a_{n} n_{j} f_{D_{0} a_{b} b_{nj} + B_{ai} a_{b} b_{nj}}^o + D_{1} a_{b} b_{nj} \right\} \, dV \]  

(3-6b)
Using the relations which express $\theta'' P$ and $\varepsilon'' P$ in terms of $q'' P$ gives

$$0 = \int V \left[ 2G_{mi} A_{am} \theta'_r D_{0} ab \varepsilon'_b + B_{ai} D_{0} ab B_{bj} q'_j P + G_{mi} A_{am} ight]$$

$$+ (D_{0} ab (2A_{bn} s n s + A_{bn} s n s) + D_{1} abc (4\varepsilon'_b \varepsilon'_c + 2\varepsilon'_b \varepsilon'_c)$$

$$+ G_{mi} A_{amn} G_{nj} j q'' P + 2G_{mi} A_{amn} \varepsilon'_n P (D_{0} ab (\varepsilon'_b + \varepsilon'_b)) dV$$

This is the second order postbuckling equilibrium equation, which may be written in the form

$$0 = K_0 q'' P + 2S' P^2_1 + P^2_1$$

where $K_0$ is again the Jacobian evaluated at the critical point, and

$$P^1_1 = \int V G_{mi} (D_{0} ab (A_{am} A_{bn} s n s + A_{am} \varepsilon'_b \varepsilon'_b + A_{am} \varepsilon'_b \varepsilon'_b)$$

$$+ 2D_{1} abc A_{am} \varepsilon'_b \varepsilon'_c ) dV$$

$$P^2_1 = \int V G_{mi} (D_{0} ab (A_{am} A_{bn} s n s + 2A_{am} \varepsilon'_b \varepsilon'_b)$$

$$+ 2D_{1} abc A_{am} \varepsilon'_b \varepsilon'_c ) dV$$

The term $S'$ in equation (3-6d) is the derivative of the fundamental path parameter with respect to the postbuckling path parameter, and occurs because of the substitutions

$$\theta'_f = \theta'_f s'$$

$$\varepsilon'_f = \varepsilon'_f s'$$

(3-6d) may be solved for the postbuckling displacement second derivatives $Q'' P$, and for the path derivative $S'$. 

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Third Order Equilibrium Equation - A third differentiation of equation (3-3) and evaluation at the critical point, gives

\[ p^{'''} = 0 = \int_V \left[ B_{ai} f_{\sigma}^P a + 3B_{ai} f_{\sigma}^P a + 3B_{ai} f_{\sigma}^P a + B_{ai} f_{\sigma}^P a \right. \\
+ B_{ai} f_{\sigma}^P a + 3B_{ai} f_{\sigma}^P a + 3B_{ai} f_{\sigma}^P a + B_{ai} f_{\sigma}^P a \\
+ B_{ai} f_{\sigma}^P a + \left. \right] dV \] (3-7a)

Substituting for \( \sigma^{'''P} \) and \( B^{'''P} \) gives

\[ 0 = \int_V \left[ 3G_{mi} A_{amn} \frac{\theta^{'''}f}{n} DO_{ab} \varepsilon^{'''}_b P + 3G_{mi} A_{amn} \frac{\theta^{'''}f}{n} \left( DO_{ab} \varepsilon^{'''}_b P \right) + Dl_{abc} \left( 4\varepsilon^{'''}_b P \varepsilon^{'''}_b P + 2\varepsilon^{'''}_b P \varepsilon^{'''}_b P \right) \right. \\
+ B_{ai} DO_{ab} \varepsilon^{'''}_b P + Dl_{abc} \left( 6\varepsilon^{'''}_b P \varepsilon^{'''}_c P + 6\varepsilon^{'''}_b P \varepsilon^{'''}_c P + 6\varepsilon^{'''}_b P \varepsilon^{'''}_c P \right) \\
+ \left. G_{mi} A_{amn} \frac{\theta^{'''}P}{a} + 3G_{mi} A_{amn} \frac{\theta^{'''}P}{n} DO_{ab} \left( \varepsilon^{'''}_b P + \varepsilon^{'''}_b P \right) + 3G_{mi} A_{amn} \frac{\theta^{'''}P}{n} \left( DO_{ab} \left( \varepsilon^{'''}_b P + \varepsilon^{'''}_b P \right) \right. \right. \\
+ Dl_{abc} \left( 2\varepsilon^{'''}_b P \varepsilon^{'''}_c P + 4\varepsilon^{'''}_b P \varepsilon^{'''}_c P + 2\varepsilon^{'''}_b P \varepsilon^{'''}_c P \right. \right. \\
+ Dl_{abc} \left( 2\varepsilon^{'''}_b P \varepsilon^{'''}_c P + 4\varepsilon^{'''}_b P \varepsilon^{'''}_c P + 2\varepsilon^{'''}_b P \varepsilon^{'''}_c P \right. \right. \left. \right) dV \] (3-7b)

Using the relations which express \( \theta^{'''P} \) and \( \varepsilon^{'''P} \) in terms of \( q^{'''P} \) gives

\[ 0 = \int_V \left[ 3G_{mi} A_{amn} \frac{\theta^{'''}f}{n} DO_{ab} \varepsilon^{'''}_b P + 3G_{mi} A_{amn} \frac{\theta^{'''}f}{n} \left( DO_{ab} \varepsilon^{'''}_b P + Dl_{abc} \left( 4\varepsilon^{'''}_b P \varepsilon^{'''}_c P \right. \right. \\
+ 2\varepsilon^{'''}_b P \varepsilon^{'''}_c P \right. \right. + B_{ai} DO_{ab} B_{bji} \theta^{'''P} + G_{mi} A_{amn} \left( DO_{ab} \left( 3A_{bns} \frac{\theta^{'''f}}{n} \theta^{'''f} \right. \right. \\
+ 3A_{bns} \frac{\theta^{'''f}}{n} \theta^{'''f} + 3G_{mi} \frac{\theta^{'''f}}{n} \left. \right) + Dl_{abc} \left( 6\varepsilon^{'''}_b P \varepsilon^{'''}_c P + 6\varepsilon^{'''}_b P \varepsilon^{'''}_c P \right. \right. \\
+ 6\varepsilon^{'''}_b P \varepsilon^{'''}_c P \right. \right. \left. \right) + G_{mi} A_{amn} G_{njq} \varepsilon^{'''}P + 3G_{mi} A_{amn} \left( DO \left( \varepsilon^{'''}_b P + \varepsilon^{'''}_b P \right) \right. \right. \left. \right) \] \( 3-8 \)
This is the third order postbuckling equilibrium equation, which may be written in the form

\[ 0 = K \delta \epsilon_{i} \delta \epsilon_{j} P + 3S''P_{21}^{3} + 3P_{3i} \]  

(3-7d)

where

\[ P_{3i} = \int_{V} G_{ni} [DO_{ab}(S' S')^{2} (A_{am} A_{bn} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) + (A_{am} A_{bn} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) + (A_{am} A_{bn} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) + (A_{am} A_{bn} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) + (A_{am} A_{bn} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} + A_{am} \epsilon_{b}^{f} \theta_{r}^{f} ) ] dV \]

The term \( S'' \) occurs in (3-7d) after making the substitutions

\[ \theta_{n}^{f} = \theta_{n}^{f} S' \]
\[ \theta_{n}^{f} = \theta_{n}^{f} S'' + \theta_{n}^{f} S'^{2} \]
\[ \epsilon_{a}^{f} = \epsilon_{a}^{f} S' \]
\[ \epsilon_{a}^{f} = \epsilon_{a}^{f} S'' + \epsilon_{a}^{f} S'^{2} \]
Equation (3-7d) may be solved for the postbuckling displacement second derivatives $Q^{''P}$, and for the path second derivative $S''$.

3.3 Solution of Postbuckling Equilibrium Equations

The postbuckling equilibrium equations (3-5d, 6d' and 7d) may be solved sequentially to yield the displacement and load derivatives necessary for construction of the postbuckling path. These equations have been formulated here for the general case of an unsymmetric $K_0$ Jacobian matrix, and the effects of nonconservative loading are discussed in Appendix B. The solution of the second and higher order equations for the unsymmetric case present some practical difficulties, however. Therefore, in contrast to the general solution outlined in Section 2 for the fundamental equations, the solution given here for the postbuckling equations will be presented for the case of a symmetric $K_0$ matrix.

First Order (Bifurcation) Solution - The first order equation (3-5d) may be solved for the eigenvector $Q^P$ of postbuckling displacements, and for the critical value of the fundamental path parameter $S$. The initial step is to relate the unknown critical displacement increments $AQ$ to the eigenvalue $S$, using the previously computed fundamental displacement derivatives:

$$Q^f_i = \dot{Q}^f_i S + \frac{1}{2} \ddot{Q}^f_i S^2$$  \hspace{1cm} (3-8)

In addition to the nonlinearity inherent in this relation, the eigenequation is nonlinear for other reasons:

1. Although increments in the displacement derivatives $\Delta \theta^f$ and displacements $\Delta Q^f$ are linearly proportional, the strain increments vary nonlinearly, i.e.

$$\Delta \varepsilon^f_a = A^*_{am} \Delta \theta^f_m + \frac{1}{2} A_{almn} \Delta \theta^f_m \Delta \theta^f_n$$

2. There are $\Delta^2$ terms ($\Delta \varepsilon^f \Delta \theta^f$, $\Delta \varepsilon^f \Delta \varepsilon^f$) in the eigenequation.
due to consideration of the nonlinear material effects
(effects of the $D_1^*$ matrix).

Because of these nonlinearities an eigensolution by direct iteration may not converge. Particular difficulty may be expected during the first few iterations, when the estimated eigenvector contains significant proportions of higher modes for which the $\Delta \theta^f$ at some locations in the structure could be much larger than the corresponding $\theta^*$. Also for such higher modes, the contribution to $\Delta Q$ by $S^2$ may be large and the $\Delta^2$ terms may be large relative to $\Delta$ terms. It is therefore necessary to solve first the linear eigen problem, obtained by dropping all nonlinear terms. When convergence has been achieved to within a specified accuracy, iteration is continued with inclusion of all terms until convergence to the desired nonlinear eigensolution.

Higher Order Solutions - With the critical point now defined by the critical value of $S$, the higher order postbuckling equations may be solved by formation and decomposition of the critical point Jacobian $K_0$. To accomplish the solution, a definition of the postbuckling path parameter $R$ is required. We here follow the general approach of Mau and Gallagher (1972) and take $R$ to be one of the postbuckling displacements, say $Q^P_m$. In the PANES program, $m$ is taken as the index of the largest component of the eigenvector $Q^P$. We then impose the requirements at every path point, $R$, that

$$R' = 1$$
$$R'' = R''' = \ldots = 0$$

(3-9)

Although the matrix $K_0$ is singular, this constraint of the $m$th degree of freedom allows the matrix to be decomposed. A somewhat different approach than this is suggested by Haftka et al. (1970), involving the introduction of an additional constraint equation to make the $K_0$ matrix effectively nonsingular. That approach

3-11
however increases the size of the matrix, and no real advantage is seen. The present approach retains the sparsity of K0.

At this point the eigenvector \( Q'P \) is again determined, using the constrained K0 matrix. This is done to achieve consistency in the calculation of \( Q'P \) and the higher order derivatives determined later, as well as for greater accuracy. In terms of the symbolic inverse \( K0^{-1} \):

\[
0 = K0_{ij}Q_{ij}^P, \text{ with } Q_{ij}^P = 1 \quad (3-10a)
\]

\[
Q_i^P = K0_{i}^{-1}(0) \quad (3-10b)
\]

The second order equation is

\[
0 = K0_{ij}Q_i'P + 2S'P2_i + P2^2, \text{ with } Q_{ij}'P = 0 \quad (3-11a)
\]

Premultiplying by \( Q_iP \), and using the symmetry of K0 with \( K0_{ij}Q_{ij}^P = 0 \), results in

\[
0 = Q_i'PK0_{ij}Q_{ij}^P + 2S'Q_i'Pp2_i + Q_i'Pp2^2 = 2S'Q_i'Pp2_i + Q_i'Pp2^2 \quad (3-11b)
\]

which then gives

\[
S^C = -Q_i'Pp2^2 / 2Q_i'Pp2_i \quad (3-11c)
\]

and \( Q_i''P = K0_{ij}Q_i''P + 3(S''P2_i + P3_i) \), with \( Q_{ij}''P = 0 \) \quad (3-11d)

The third order equation is

\[
0 = K0_{ij}Q_i'''P + 3(S''P2_i + P3_i), \text{ with } Q_{ij}'''P = 0 \quad (3-12a)
\]

Premultiplying by \( Q_iP \) as before, results in

\[
0 = S''Q_i'Pp2_i + Q_i'Pp3_i \quad (3-12b)
\]

3-12
which then gives

\[ S'' = -Q'_{\text{P}}P_{3_{\text{P}}} / Q'_{\text{P}}P_{2_{\text{P}}} \]  

(3-12c)

and \[ Q''_{\text{P}} = K_{0_{\text{ij}}}^{-1}(-3S''P_{2_{\text{P}}} - 3P_{3_{\text{P}}}) \equiv -3(S''Q_{2_{\text{P}}} + 3Q_{3_{\text{P}}}) \]  

(3-12d)

With the critical point derivatives of \( S \) and \( Q^P \) known, the post-buckling path can be constructed. The variation of load parameter \( \Lambda \) with postbuckling path \( R \), is defined by

\[
\begin{align*}
\Lambda' &= \dot{\Lambda}S' \\
\Lambda'' &= \ddot{\Lambda}S'' + \dot{\Lambda}S'^2
\end{align*}
\]

(3-13)
4.0 FINITE ELEMENT PROGRAM AND NONLINEAR SOLUTION ROUTINES

4.1 General Program Characteristics

The major goal of this research effort was the development of improved nonlinear solution techniques and subroutines. It was decided that the most effective way of accomplishing this goal was to develop a practical nonlinear finite element program, into which the various subroutines could be incorporated for checkout and verification. This has been accomplished, and the resulting finite element program has been given the acronym PANES (Program for Analysis of Nonlinear Equilibrium and Stability). Although PANES is a pilot program and is by no means a general structural analyzer (it utilizes only the constant strain triangle element, for 2-D in-plane or 3-D membrane analysis) it demonstrates all of the basic techniques and operations necessary for nonlinear analysis by more general types of finite elements. The program handles geometric nonlinearities and arbitrary nonlinear elastic materials (including very large strain cases), as well as certain forms of nonconservative loadings, i.e. those due to follower-force pressure loadings where the surfaces change in size and orientation. Extension of the program to cases of inelastic materials is considered to be relatively straightforward, with the introduction of appropriate stress-strain constitutive relations.

The present pilot version of PANES has three basic capabilities:

1. Analysis of nonlinear structures without critical points, i.e. tracing of simple nonlinear equilibrium paths under a specified general (non-proportional) loading. Various solution techniques are available, with automatic calculation of load step sizes.

2. Traversing of limit (maximum and minimum) type critical points, with automatic continuation of the load-path history.
3. Determination of bifurcation type critical points, and prediction of the postbuckling behavior and direction of travel, by means of path derivatives computed at the bifurcation point. Automatic switching from the fundamental path to a postbuckling path, and continuation along the postbuckling path, have not yet been included. Thus the postbuckling capability should be regarded as still in a developmental stage.

4.2 PANES Nonlinear Analysis Routines

This section describes briefly the purpose and capabilities of the program subroutines, in the order in which they appear in the PANES program. Some of these are basic finite element routines, while others are specialized routines needed for generating and solving nonlinear structural equations.

**BIGS** - Initializes program variables (serves as the calling subroutine for most of the input data reading routines). Also provides problem restart capability by reading or writing the restart tape.

**READRS** - Reads data file numbers and start or restart codes.

**READO** - Reads problem identification title. Also reads incremental and iterative constants, such as those relating to the predictor and corrector types, the finite difference expansions for nonlinear materials, and the techniques for continuation of the equilibrium path through limit points.

**READ1** - Reads basic structural codes and values, and material constants for each material.

**READC** - Reads user-defined special nodal coordinate systems.
READM - Reads mesh data, including nodal locations and coordinate system codes, and element data.

READK - Reads codes to determine degrees of freedom with specified forces, displacements or constraints.

READP - Reads two load reference curves which define distribution of the applied generalized nodal loads.

READPR - Reads a pressure load reference curve which defines the distribution of the applied pressure loads (one intensity for each element).

READI - Reads incremental load data, including the nodal load and pressure load curve factors for the total load at the end of each increment.

HEAD - Writes a heading output for each load increment step, including load parameter value, number of iterations required and accuracy achieved.

OUTLIM - Predicts and outputs limit point values for the load parameter, and nodal forces and displacements.

OUTPQ - Outputs nodal forces and displacements.

OUTE - Outputs element strains.

QFILL - Uses vector of system-level nodal displacements Q to form vector of element-level nodal displacements q for an element.

PFILL - Takes vector of element-level nodal forces p and adds them to system-level force vector P.
DFILL - Uses element nodal displacement vector \( q \) to compute vector of displacement derivatives \( \theta \) within the element.

EFILL - Uses element displacement derivatives vector \( \theta \) to compute vector of strains \( \varepsilon \) within the element.

AFILL - Uses element displacement derivatives \( \theta \) to form Lagrangian AO or Al matrix within the element.

GFILL - Uses element displacement functions to form the \( \theta-q \) transformation matrix \( G \).

MTRAN - Matrix transformation routine, which performs operations of the type \( K_{ij} = D_{ab}B_{ai}B_{bj} \) for given \( D \) and \( B \) matrices.

ROTQ - Transforms element displacements or forces, from either nodal to element or element to nodal coordinate system.

ROTK - Transforms element stiffness matrix from element to nodal coordinate system.

FORCE - Computes internal nodal generalized forces corresponding to given nodal displacements.

PFORCE - Computes applied nodal force loadings, using nodal load reference curves and corresponding load factors.

EFORCE - Computes nodal forces due to applied pressure loadings, using pressure reference curve and factor, and the current area and orientation of each element (determined from geometry and current displacements).

ERCOMP - Computes and outputs error norm for each residual force iteration, using applied (external) forces and computed (internal) forces.
**STRAIN** - Computes strains for each element using geometry and nodal displacements.

**ENERGY** - Evaluates the strain energy density for an element at given strain components. This routine will in general be a user supplied routine based on the types of materials being used in the structure.

**EVAL** - Performs the function (strain energy) evaluations at the current strain state, and at the required adjacent "perturbed" states necessary to establish a strain energy expansion in terms of incremental strains. EVAL calls the STRAIN routine for evaluation, and defines the evaluation points by using the user-specified finite difference sizes. A first, second or third order expansion may be specified, and the corresponding function evaluations are returned in the form of a vector.

**U2FORM** - Forms coefficients for a general second order Taylor series expansion, using function values provided by EVAL. Used to develop the strain-energy related tensors \( \sigma_i \) and \( D\sigma_{ij} \) for a material at current deformation state.

**U3FORM** - Similar to U2FORM, but forms coefficients for a general third order expansion. Develops the tensors \( \sigma_i \), \( D\sigma_{ij} \) and \( D^3_{ijk} \).

**UFILL** - Calling routine which calls either U2FORM or U3FORM, depending on desired expansion order.

**CFORM** - Forms the contribution to the Jacobian stiffness matrix due to the nonconservative pressure loadings.

**GENER8** - Generates the elemental Jacobian matrix using the current geometry and the material tensors \( \sigma_i \) and \( D\sigma_{ij} \). Also adds contributions from CFORM if loading is nonconservative.
USUM1 - Performs a summing operation between a second or third order tensor function and its vector argument, to give a vector.

USUM21 - Performs a summing operation between a third order tensor function and its two (different) vector arguments, to give a vector.

P1COMP - Computes the nonlinear load term P1*, required in generating the second order fundamental equilibrium equations.

RATES - Computes the first and second order fundamental load parameter and displacement rates.

STEP - Provides automatic calculation of a fundamental path load step size, and techniques for traversing limit points.

EIGEN1 - Computes the pseudo force term P1^1, for use in the inverse power iteration eigensolution process.

EIGEN - Eigensolution routine for inverse power iteration. Calls EIGEN1 routine.

POST2 - Computes the second order postbuckling pseudo force terms P2^1 or P2^2.

POST3 - Computes the third order postbuckling pseudo force term P3.

PRATES - Computes the first and second order postbuckling load and displacement rates, and third order displacement rate, at the bifurcation point.

VDOT, VCROSS, VLENTH, VNORM - Vector subroutines for computing dot product, cross product, length, and normalizing a vector, respectively.
MERGE - Merges elemental Jacobians into system Jacobian, with provision for constrained degrees of freedom. Forms general unsymmetric Jacobian matrix.

DECOMP - Decomposes unsymmetric Jacobian using Gauss wavefront type procedure. Takes advantage of sparsity but uses total square matrix for storage without packing or external storage devises. (This is a small pilot version decomposition routine.)

SOLVE - Performs forward and backward substitution for unsymmetric Jacobian matrix to provide solution vectors.
4.3 Summary of PANES Input Data

A pictorial of the PANES input deck is shown in Figure 4-1. The input data consists of the following three general types:

Type C: Data on the usual card file. These are data which are needed for each start or restart.

Type I: Data on File I. These are basic structural data for a given problem, such as material properties and mesh data. They are the same for all load increments and are needed only when starting.

Type II: Data on File II. File II is not used in the current PANES version. It is provided for possible future use as a file of incremental data (e.g. additional nodal and thermal load data).

The data included on each file are described below. Formats are consistent with FORTRAN IV conventions.

C-1. Start-restart code and data file numbers:

a. "START" if new problem, or "RESTART" if restarting.

b. If starting give unit number for file I.

c. Unit number for file II (need not be given).

d. Unit number for output file (e.g. printer).
Figure 4-1: PANES INPUT DECK SETUP

NOTES:

C IS CARD FILE
I, II ARE CARD OR TAPE FILES
DATA ON FILE I IS USED ONLY WHEN STARTING A NEW PROBLEM.
DATA ON FILE C IS USED WHEN STARTING OR RESTARTING
FILE II IS RESERVED FOR POSSIBLE FUTURE USE
e. If restarting give load increment number from the end of which a restart is to be made.

f. If restarting give input restart-tape unit number.

g. If data is to be saved for future restart give output restart-tape unit number.

Format (A4, 6X, 6I5)


Format (20A4).

C-3. Program control constants (any constant left blank is assigned a default value):

a. Specified order of material incremental stress-strain expansion to be used (2 is exact for linear material, maximum order is 3). Default order is 3.

b. Solution predictor type. Type 1 = 1st order, Type 2 = 2nd order.

Default type = 2.

c. Maximum number of Jacobian updates per load increment step.

Default = 0.

d. Maximum number of residual force iterations per Jacobian update.

Default = 5.
e. Maximum allowable residual force error norm.

Default = $1 \times 10^{-8}$.

Format (4I10, F10.0)

C-4. Perturbation difference magnitudes for evaluating strain energy.

a. Difference for computing stiffnesses.

Default = $1 \times 10^{-3}$.

b. Difference for computing forces.

Default = $1 \times 10^{-8}$.

Format (2F10.0)

C-5. Program control constants

a. Number of increment subdivisions to be performed as load nears a limit value. Default = 3.

b. Ratio of limit load to load increment values at which limit point is to be traversed. Default = 0.1.

c. Increment step size limitation, computed from slope of load parameter versus path parameter curve, and equal to change in slope divided by average slope.

Default = 0.5.
d. Maximum load increment step size (used especially in unloading), and defined as a factor times the specified load increment.

Default = 1.0.

e. Maximum fraction of current load increment by which load is allowed to reduce after passing a maximum limit point.

Format (I10, 4F10.0)

I-1. Basic structure definition

a. Code for element pressure loads. Code 0 = no pressures, Code 1 = pressures. Default code is 0.

b. Degree of freedom per node (2 or 3). No default value.

c. Default thickness for all elements.

Format (2I10, F10.0)

I-2. Material property definitions. For each material give material I.D. number, and 2 material constants for use by the strain energy evaluation routine.

Format (I10, 2F10.0)

Blank card after data for last material.
I-3. For each special Cartesian coordinate system: the identification number (integer > 2) and counter-clockwise angle (degrees) from basic system X-axis to special system x-axis.

Format (I10, F10.0)

Blank card after last coordinate system.

I-4. For each node: Node number; identification number of coordinate system to define location; X, Y and Z (or R, \( \theta \) and Z); identification number of coordinate system to define displacements. (Coordinate I.D. number 0 implies the basic Cartesian system, 1 implies the basic cylindrical system).

Format (2I5, 3F10.0, I5)

Blank card after last node.

I-5. For each element: element number, material number, thickness, three node numbers (counter-clockwise order).

If thickness is left blank, default value from I-1c is used.

Format (2I5, F10.0, 3I5)

Blank card after last element.

I-6. For each DOF with specified displacement or constraint:

If specified displacement, give node number and component (1, 2 or 3) number;

4-13
If specified constraint, give node and component number, and independent node and component to which DOF is constrained (independent component number is + for specified force, - for specified displacement). User has option of from 1 to 4 values per card.

Format (4(4I5))


Number of vectors (for current program version must be 2)

Format (I10)

For each nonzero component of load vector:
node number, component number (1 = X or R, 2 = Y or θ, 3 = Z), force or displacement value. User has option of from 1 to 4 values per card.

Format (4(2I5, F10.0))

Blank card after last value of each vector.

I-8. Pressure Load Reference Vector. (Input only if pressure code in data item I-1 is nonzero.

Number of vectors (for current program version must be 1)

Format (I10)

For each nonzero component of pressure load vector:
element number, pressure intensity. User has option of from 1 to 4 values per card.

Format (4(I10, F10.0))

Blank card after last value of vector.
C-6. Incremental load data

Number of load increments

Format (I10)

For each load increment: solution predictor type (if left blank, value from C-3b is used), the cumulative factors to be applied to the nodal load reference vectors, pressure value for all elements. Pressure is applied in element positive z-coordinate direction.

Format (I10, 3F10.0)

C-7. Final blank card.

Problems may be run consecutively (first data item for each problem follows immediately after last item of preceding problem). Final blank card indicates that all problems have been read.

4.4 Summary of PANES Output

The description of PANES output is conveniently divided into two parts. The first is primarily an echo check of the input data, and the second part consists of output results for each load increment.

4.4.1 Echo Check of Input Data

Initial Output - The first page of PANES output for a problem is essentially an echo check of input items C-1 to C-5, I-1 and I-2. An indication is given as to whether the problem is being started or restarted. If it is restarted then the previous increment number is given, from the end of which the restart is
progressing. Next the problem I.D. title is printed, followed by the various control constants and finite difference magnitudes (DFE and DFF). The limit point related control constants (MJUMP, JUMPR, SLOPED, FLAMAX and LAMIN) are then printed. Finally the basic structural quantities from I-1, and the material property constants from I-2 are printed.

Special Coordinate Systems - These are the user-defined direction (special Cartesian) systems of input data item I-3. Quantities printed are the system number, and counter-clockwise angle (in degrees) from the basic X axis to the special-system X axis.

Node Definitions - The information given in input item I-4 is printed. Values are the node number, location coordinate system number (0 = basic Cartesian, 1 = basic cylindrical), X or R coordinate, Y or θ (degrees) coordinate, Z coordinate, direction coordinate system number (0 = basic Cartesian, 1 = basic cylindrical, >1 = number of special user defined system).

Element Definitions - The information given in input item I-5 is printed. Values are the element number, material I.D. number, element thickness, the three element node numbers in counter-clockwise order, and the computed element area.

Force-Displacement-Constraint Prescriptions - These are the codes given in input data item I-6. Quantities printed are the dependent node and component number, and independent node and component number. (If specified displacement, no independent numbers are given).

Nodal Load Reference Vectors - For each input component of the two load vectors from input item I-7, the node number, component number, and load value are printed.
Pressure Load Reference Vector - For each input component of the pressure load vector from input item I-8, the element number and pressure intensity are printed.

Incremental Load Data - Quantities related to input data item C-6 are printed. First is printed the number of load increments to be run. Then for each increment is given the increment number, input or default value for the predictor type, and factors to be applied to the two nodal load reference vectors and the pressure load reference vector.

4.4.2 Results for Each Load Increment

Interactive Error Values - An error norm computed at the end of each iteration is printed. The error norm is obtained by a ratio of unbalanced (residual) forces to total forces.

Increment Heading - The load increment and load step numbers are printed, along with the load increment and load step values at the end of the step. Following this are the nodal load reference vector factors, the element pressure vector factor, the predictor type for the increment, the maximum allowable number of Jacobian updates and the number performed during this load step, the maximum allowable number of iterations per update and the number performed since the last update, and the maximum allowable error norm and the error norm actually achieved.

Forces and Displacements - The cumulative nodal displacements and corresponding internal forces are output. The node number is printed, followed by the U, V and W (or R, θ, Z) components of force and displacement.

Strains - The cumulative element strains are output. The element number is printed, followed by the XX, YY, and XY strains in the element coordinate system.
Limit Point Output - When a limit point is traversed, the predicted value of the incremental limit load parameter is output, followed by the predicted limit forces and displacements, and strains.

Bifurcation Point Output - When an eigenvalue solution is performed to determine a critical point, the eigenvalue computed for each inverse power iteration is printed, along with the location of the maximum value in the eigenvector.

Decomposition Output - Whenever the Gauss decomposition routine is called, it prints the value (sign and base 10 logarithm) of the Jacobian stiffness determinant.
5.0 ILLUSTRATIVE PROBLEMS

Four example problems are presented here in order to illustrate various aspects of nonlinear equilibrium and stability theory, and to demonstrate use of the developed nonlinear subroutines and the PANES finite element program. Section 5.1 describes a snap-through truss problem with geometric nonlinearity and a maximum and a minimum limit point. Section 5.2 describes a simple pressure membrane with nonconservative type loading (changing load area), resulting in a maximum limit point. The toroidal membrane in Section 5.3 is a fairly difficult problem, involving nonlinear (Mooney) material with follower-force loads and changing load areas. It results in very large displacements and strains, and a maximum and a minimum limit point. This problem demonstrates some unique capabilities of the PANES program. Finally, Section 5.4 describes a simple bifurcation/postbuckling model, with asymmetric behavior.

5.1 Snap-Through Truss

This is a problem similar to that used as a test case by several researchers in nonlinear structural analysis, e.g., Haisler et al. (1971). The system consists of a single inclined bar (or one half of a symmetric two-member truss) as shown in Figure 5-1. The bar has length 1.0 with axial stiffness \( AE = 2 \times 10^7 \) (Hookean material), is inclined initially at a slope of 1:100, and is subjected to a vertical end load \( P \). The PANES program idealization of this system used two constant strain triangle elements, with modulas \( E = 2 \times 10^8 \) and thickness \( = 0.1 \). Node 4 was constrained to have vertical displacement equal to that at node 2, so that a system with essentially one degree of freedom (vertical displacement \( Q \)) is obtained.

The expression for the axial strain, \( \varepsilon \), is given by

\[
\varepsilon = -.01Q + 0.5Q^2
\]  
\( (5.1-1a) \)
LIMIT $P = 3.85$, $Q = 0.00423$

LIMIT $P = -3.85$, $Q = 0.01577$

Figure 5-1: SNAP-THROUGH TRUSS
and stress, $\sigma$, is given by

$$\sigma = -2,000,000Q + 100,000,000Q^2$$  \hspace{1cm} (5.1-1b)

Thus the axial force is $-200,000Q + 10,000,000Q^2$, from which it may be shown that the vertical applied force, $P$, is given by the following basic equilibrium equation.

$$P = 2000Q - 300,000Q^2 + 10,000,000Q^3$$  \hspace{1cm} (5.1-2)

Differentiating with respect to $Q$, and evaluating at a reference equilibrium configuration ($Q^*, P^*$), gives the first order equilibrium equation

$$\dot{P}^* = 2000\dot{Q}^* - 600,000Q^*\dot{Q}^* + 30,000,000Q^*Q^{**}$$  \hspace{1cm} (5.1-3a)

or

$$\ddot{P}^* = K_0\dot{Q}^*$$  \hspace{1cm} (5.1-3b)

where the Jacobian stiffness is

$$K_0^* = 2000 - 600,000Q^* + 30,000,000Q^*^2$$

A second differentiation and evaluation of equation (5.1-2) results in the second order equilibrium equation

$$\ddot{P}^* = K_0\dot{Q}^* + P_1^*$$  \hspace{1cm} (5.1-4)

where the pseudo force $P_1$ is defined by

$$P_1^* = 600,000\dot{Q}^*^2 + 60,000,000Q^*\dot{Q}^*^2$$

Using the equilibrium equations (5.1-3) and (5.1-4), the $P$-$Q$ path history can be computed by various incremental and iterative approaches, including identification of limit points.

5-3
(Of course for this simple one-degree-of-freedom system, the path can be obtained immediately from the basic equilibrium equation (5.1-2)). The P-Q path history is shown in Figure 5-1. The PANES program solution was accomplished using six user-specified load increments \( P = 1.0, 2.0, 3.0, 4.0, 5.0, 10.0 \), although about 30 additional load steps were selected automatically (mostly to achieve the desired accuracy in locating and traversing the limit point regions). Most load steps required only one or two residual force iterations, with use of a second order predictor.

5.2 Simple Pressure Membrane

One of the simplest problems which can be used to illustrate some of the effects of nonconservative loading in stability analysis is the simple pressure membrane shown in Figure 5-2. The system consists of a flat membrane 2.0 wide by 1.0 high with unit thickness, and subjected to a uniform pressure intensity \( \lambda \) on one side. The ends of the membrane slide along the \( 45^\circ \) supports, and are constrained to move together equally in the X direction. This gives a single degree-of-freedom system, with X-direction force \( P \) and displacement \( Q \), and the membrane undergoes a uniform stretching in the Y direction. The solution was verified by a finite element analysis using the PANES program. The finite element mesh consisted of the two constant strain triangle elements shown in Figure 5-2, with X-direction displacements at nodes 2-4 constrained to equal the displacement at node 1. Zero displacements were enforced in the Z direction.

Considering large displacement and large strain effects, the stretch in the Y direction is denoted by \( \lambda \), and is equal to the change in length divided by original length. The strain-energy density \( U \), measured per unit undeformed volume, is taken to be defined by the function

\[
U = C_1(\lambda-1)^2 + C_2(\lambda-1)^4 = C_1Q^2 + C_2Q^4 \quad (5.2-1)
\]
Figure 5-2: SIMPLE PRESSURE MEMBRANE

LIMIT POINT
Q = 1.0, ω = 8.0
Note that we could define a Lagrangian strain $\varepsilon$, and stress-like quantity $\sigma$, by

$$\varepsilon = \frac{1}{2}(\lambda^2 - 1)$$

$$\sigma = \frac{3}{2} \frac{U}{\frac{Q}{\lambda}} = \frac{3}{2} \frac{U}{Q} = 2C_1Q + 4C_2Q^3$$

However, this is not necessary in the present problem, for which the force can be derived directly from the strain-energy function.

The basic equilibrium equation for the system is written using the equivalence of external force (defined in terms of the pressure loading $\lambda$) and internal force (defined as the derivative of strain energy with respect to displacement $Q$):

$$P = 2\lambda(1+Q) = 2(2C_1Q + 4C_2Q^3) \quad (5.2-2)$$

Choosing specific values for the material constants $C_1 = 10$ and $C_2 = -1$, the basic equilibrium equation may be written as

$$\lambda(1+Q) = 20Q - 4Q^3 \quad (5.2-3)$$

Differentiating equation (5.2-3) provides the first order equilibrium equation

$$\dot{\lambda}(1+Q) + \lambda \dot{Q} = 20 \dot{Q} - 12Q^2 \ddot{Q} \quad (5.2-4a)$$

or at an equilibrium configuration ($\lambda^*, Q^*$) we can write

$$\dot{\lambda}^* = (20 - 12Q^*^2 - \lambda^*)/(1+Q^*) \dot{Q}^* \quad (5.2-4b)$$

5-6
From this equation it can be seen that the value of a Jacobian stiffness $K_0^*$, relating $\dot{\Lambda}^*$ with $\dot{Q}^*$, is

$$K_0^* = \frac{(20 - 12Q^*^2 - \Lambda^*)}{(1+Q^*)} \quad (5.2-4c)$$

(For simplicity we have here defined the Jacobian relative to $\dot{\Lambda}$ rather than $\dot{P}$).

Differentiating equation (5.2-3) a second time gives the second order equilibrium equation, as

$$\ddot{\Lambda} (1+Q) + 2\dot{\Lambda} \dot{Q} + \Lambda \ddot{Q} = 20\ddot{Q} - 24\dot{Q}^2 - 12Q^2 \dot{Q} \quad (5.2-5a)$$

or at an equilibrium configuration we can write

$$\ddot{\Lambda}^* = K_0^* \ddot{Q}^* + P_1^* \quad (5.2-5b)$$

where the pseudo force $P_1$ is given by

$$P_1^* = \frac{(-24Q^*\dot{Q}^*^2 - 2\dot{\Lambda}^*\dot{Q}^*)}{(1+Q^*)} \quad (5.2-5c)$$

To illustrate the actual behavior of the system, we now choose $Q$ as the path parameter, and without loss of generality specify at every point the conditions $\dot{Q} = 1$ and $\ddot{Q} = 0$. To aid in determination of a limit type critical point, we have the condition $\dot{\Lambda}^* = 0$ at the limit point. Using this condition, and solving equation (5.2-4b) with the use of (5.2-3), we find a limit point at ($\Lambda^* = 8.0$, $Q^* = 1.0$). The entire $\Lambda-Q$ path history may be determined by various incremental and iterative predictor-corrector schemes, and is shown in Figure 5-2. The PANES program solution to this problem used a second-order predictor with residual-force corrective iterations.
5.3 Toroidal Membrane

The problem illustrated here is a toroidal membrane under internal pressure, shown in Figure 5-3. This structure exhibits a highly nonlinear type of behavior with very large displacements and strains, and both a maximum and a minimum limit point. The second-order predictor was employed in a PANES program solution, along with residual-force corrective iterations to achieve equilibrium at each load step. Both the predictor and corrector incorporated all changing load area and follower-force effects. (This gave an unsymmetric $K_0*$ matrix, whose decomposition was obtained by the Gauss wavefront procedure. It was apparent that the unsymmetric effects became large enough that their inclusion was necessary for convergence).

The torus was assumed to be of Mooney material with constants $C_1 = 80$ and $C_2 = 20$, and was analyzed using plane-stress membrane elements. Geometry and displacement components are defined in Figure 5-3. The torus has major radius 10, minor radius 2, and thickness .05. Cylindrical coordinates were employed to model a wedge-shaped segment of the major circumference, of from 2 to 10 degrees arc. Constraints were employed in the radial and vertical directions in order to equalize corresponding displacements along the two sides of the wedge.

Table 5-1 summarizes computed values of key displacements for the user-defined (input) pressure increments and the computed limit pressures, obtained with three different meshes. (N denotes the number of subdivisions over one half of the minor circumference). The indicated convergence with mesh refinement is of the kind to be expected, with finer meshes giving a more flexible structure, and resulting generally in somewhat larger displacements and lower limit point values. Computer run times ranged from 1 minute (IBM central processor time) for mesh $N = 4$, to 7 minutes for mesh $N = 12$. These times should be regarded in a
qualitative fashion only, since for example much of the time was spent in solving the linear unsymmetric stiffness equations and this time could be reduced by use of a production type equation solver.

More detailed results for the fine mesh \((N = 12)\) are shown in Table 5-2. There the data columns represent respectively the pressure, load increment number (user-specified increment), step number (where the PANES program automatically divided the specified increment into a number of smaller steps), number of residual-force type iterations performed in order to achieve the required accuracy, and values of the radial and vertical displacements at key points. Figure 5-3 gives very interesting plots of two key load-displacement paths, and indicates that no difficulties were caused by a displacement which followed an extremely irregular "doubling back" type of path, including sharp curvature sections. The basic results for this problem are corroborated by another solution to the same problem by Key (1974), who developed a finite element program with a Newton Raphson solution technique, and obtained results for pressure levels up to near the first limit point.

It may be of use for comparison/test purposes to mention corresponding results obtained by increasing the major radius from 10 to 12. Maximum and minimum limit points occurred at pressures, \(h\), of 4.355 and 4.125, respectively, while maximum displacements (at \(h = 5.0\)) increased roughly 20 percent.
Figure 5-3: TORUS SOLUTION RESULTS
<table>
<thead>
<tr>
<th>$h$</th>
<th>$U_A$</th>
<th>$U_B$</th>
<th>$W_B$</th>
<th>$U_C$</th>
</tr>
</thead>
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<td>-0.0015</td>
<td>0.0120</td>
<td>0.0076</td>
</tr>
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<td>0.0484</td>
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<td>0.1080</td>
<td>0.1130</td>
</tr>
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</tr>
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</tr>
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<td>1.438</td>
<td>1.203</td>
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**Limit points**

5-11
## TABLE 5-2: TORUS INCREMENTAL RESULTS (N = 12)

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† Input pressure loads. Intermediate pressures were selected automatically by the program.

†† Limit points.
5.4 Asymmetric Buckling Model

The problem illustrated here is that used by several investigators of bifurcation and postbuckling behavior, e.g., Thompson (1970). The model consists of a spring and rigid bar as shown in Figure 5-4. The asymmetric postbuckling behavior is due to the decreasing resisting moment arm of the spring force about point 0, as the top of the bar deflects to the right.

The conservative load A is applied vertically to the top of the bar. The spring is initially inclined at 45 degrees, and has constant stiffness K. This is a single degree of freedom system, defined by the horizontal displacement Q. The vertical component of distance from O to B at any time is equal to \( \sqrt{1-Q^2} \), and the horizontal distance from A to B is 1+Q, from which the length of the spring is found to be \( \sqrt{2(1+Q)} \). The moment arm of the spring force about point 0 is then determined as \( \sqrt{1-(1+Q)/2} \). Equating the external applied and internal resisting moments gives the basic postbuckling equilibrium equation for the system as

\[
Q = K(\sqrt{2(1+Q)} - \sqrt{2}) \sqrt{1-(1+Q)/2} = K(\sqrt{1+Q} - 1) \sqrt{1-Q}
\]  \hspace{1cm} (5.4-1a)

or

\[
A = K(\sqrt{1-Q^2} - \sqrt{1-Q})/Q
\]  \hspace{1cm} (5.4-1b)

Evaluating A from this expression using a small finite difference in Q, gives the critical load value as

\[
A = \frac{1}{2} K
\]  \hspace{1cm} (5.4-2)
Similarly a second-order finite difference evaluation gives the critical asymmetric load rate as

$$\frac{\partial A}{\partial Q} = -\frac{3}{8} K$$  \hspace{1cm} (5.4-3)

The location of the critical (bifurcation) point was verified by a PANES program solution.

![Diagram of asymmetric buckling model](image)
6.0 CONCLUSIONS AND RECOMMENDATIONS

Conclusions - The present work provides improved techniques for solution of structures with material and geometric nonlinearities. FORTRAN subroutines have been developed, and incorporated into a new nonlinear finite-element program called PANES (Program for Analysis of Nonlinear Equilibrium and Stability). A new approach is developed for representing an arbitrary nonlinear material in terms of a finite-difference generated stress-strain expansion, and is considered to be of major significance. This approach leads to formulation of perturbation-type equilibrium equations of any desired order, and is effective even for numerically integrated finite elements with large degrees of freedom. The formulation should provide a unifying basis for design of many nonlinear structural analysis programs.

The present PANES program is a pilot version, capable of analyzing problems with large strain and arbitrary nonlinear elastic materials, and provides membrane finite elements (two or three degrees of freedom per node) with nonconservative pressure loading. It includes automated techniques which have been developed for selection of load step sizes, and for locating and traversing maximum and minimum limit-type critical points. Subroutines are also included for location of bifurcation-type critical points on a general nonlinear prebuckling path, and for determining the symmetric or asymmetric postbuckling behavior. The postbuckling capabilities have not yet been completely automated and tested, however, and should be regarded as being in a developmental stage.

Recommendations - The PANES program solution routines provide a significant pilot capability for analysis of structures with highly nonlinear material and geometric effects, and should now
be extended and evaluated for a wider class of practical structures. Listed below are recommendations for future work.

1. The first priority should be given to improving and verifying the PANES postbuckling subroutines. This effort should include the generalization of program logic to accept additional types of finite elements, and in particular the addition of simple bar elements to the program. These elements will simplify the study of postbuckling results, and are desirable for initial verification because of the somewhat complex nature of the new nonlinear material postbuckling theory. Automated techniques should be incorporated for branching to the postbuckling path, similar to the existing PANES techniques for traversing limit points.

2. The program should be extended to incorporate a number of higher-order finite elements. Important candidate elements are plates, shells, and isoparametric-solid elements. Such elements will greatly extend the analysis capabilities of the program, and also importantly demonstrate the effectiveness of the new nonlinear solution techniques for elements which are numerically integrated and have large degrees of freedom.

3. PANES now handles an arbitrary nonlinear elastic material, by use of the proper material strain-energy definition. Formation of the stress-strain expansion relation should be generalized to cases of inelastic material, i.e., those materials for which a strain-energy function does not exist. The concept of this generalization is not difficult, but some study is required to develop an effective algorithm for forming the higher-order stress-strain expansion terms.

4. A number of largely theoretical improvements should be studied. These include the treatment of multiple and closely-spaced critical points (as often occur in an optimally designed light-weight structure), and the method
of postbuckling behavior solution for cases of an
unsymmetric Jacobian stiffness matrix. Incorporation of
a third-order fundamental path predictor also appears
desireable, especially for use in prediction of bifurcation-
type critical points. The nonlinear eigenvalue solution
for these points is somewhat costly, and would have to
be performed less often with the higher-order predictor.

5. The program size capability should be increased to handle
the expected range of practical nonlinear structural
problems. This involves some reorganization of the main
program logic, and the addition of a production-type
linear equation solver such as the Gauss-wavefront routines
used in the BOPACE elastic-plastic-creep program (Vos and
Armstrong (1973)).
7.0 REFERENCES AND BIBLIOGRAPHY


7-1


APPENDIX A: FINITE DIFFERENCE EXPANSIONS

For the type of nonlinear solution techniques utilized in this work, it is necessary to generate the Jacobian stiffness matrix as well as various force-type vectors associated with residuals and nonlinear predictor quantities. In the general case involving nonlinear materials, these quantities cannot be effectively determined by an explicit process, and must be generated numerically. The numerical representation may be based on a direct expansion of the generalized forces in terms of displacements (e.g. the method for Jacobian generation used by Oden and Key (1970)), or, as in the approach used here, it may be based on an expansion of stresses in terms of strains. In any case the procedure requires the expansion of a dependent function of several independent variables, about a known reference point.

An effective expansion procedure has been developed in the present work by use of a Taylor series, in which the expansion coefficients (partial derivatives) are evaluated using finite difference expressions. After a study of various alternatives, it was concluded that the most efficient scheme involves forward differences rather than central differences, because the forward differences result in simple formulas and require a minimum number of function evaluations. In addition, if an approximate solution path increment is known, i.e. if the approximate increments which will occur in the independent variables are known, then a more accurate function representation can be obtained with the forward difference scheme by selecting the appropriate difference values. Difference coefficients are derived here for expansions of linear, quadratic and cubic form.

**Linear Form** - Coefficients for a linear expansion correspond to those in a two-point forward difference formula. The derivation is rather trivial, but it serves to illustrate the basic procedure.
The Taylor series expansion of an arbitrary function $f$, in terms of independent variables $x_i$, is

$$f = f^1 + f_i \Delta x_i$$  \hspace{1cm} \text{(A-1)}$$

where $f_i = \partial f / \partial x_i$ denotes the partial derivative of $f$ with respect to the $i$th independent variable, and $\Delta$ denotes an incremental quantity. The unique types of terms may be derived by considering only one of the independent variables, which we denote simply by $x$. Referring to Figure A-1, we describe the values of $f$ at points 1 and 2 by the linear expansion

$$\begin{bmatrix} f^1 \\ f^2 \end{bmatrix} = \begin{bmatrix} 1 & 0 \\ 1 & 1 \end{bmatrix} \begin{bmatrix} f^1 \\ f_x \Delta x \end{bmatrix}$$  \hspace{1cm} \text{(A-2)}$$

Inversion of this relation gives explicit definition to the difference coefficients, in the form of the matrix in the inverse relation:

$$\begin{bmatrix} f^1 \\ f_x \Delta x \end{bmatrix} = \begin{bmatrix} 1 & 0 \\ -1 & 1 \end{bmatrix} \begin{bmatrix} f^1 \\ f^2 \end{bmatrix}$$  \hspace{1cm} \text{(A-3)}$$

or

$$f_x = (-f^1 + f^2) / \Delta x$$  \hspace{1cm} \text{(A-4)}$$

![Figure A-1: Linear Difference Expansion](image-url)
Quadratic Form - Coefficients for the quadratic expansion correspond to those in a three-point forward difference formula. Because the expansion involves terms no higher than second order, the unique types of difference coefficients can be derived by considering only two of the independent variables, say \( x \) and \( y \). The corresponding Taylor series expansion for the function \( f(x,y) \) is given by the expression

\[
f = f^1 + f^\Delta x + f^\Delta y + \frac{1}{2} f^{xx} (\Delta x)^2 + f^{yx} \Delta y \Delta x + \frac{1}{2} f^{yy} (\Delta y)^2 \quad (A-5)
\]

Figure A-2: Quadratic Difference Expansion

Referring to Figure A-2, we write

\[
\begin{bmatrix}
  f^1 \\
  f^2 \\
  f^3 \\
  f^4 \\
  f^5 \\
  f^6 \\
\end{bmatrix} = \begin{bmatrix}
  1 & 0 & 0 & 0 & 0 & 0 \\
  1 & 1 & 0 & 1/2 & 0 & 0 \\
  1 & 0 & 1 & 0 & 0 & 1/2 \\
  1 & 2 & 0 & 2 & 0 & 0 \\
  1 & 1 & 1 & 1/2 & 1 & 1/2 \\
  1 & 0 & 2 & 0 & 0 & 2 \\
\end{bmatrix} \begin{bmatrix}
  f^1 \\
  f^\Delta x \\
  f^\Delta y \\
  f^{xx} (\Delta x)^2 \\
  f^{yx} \Delta y \Delta x \\
  f^{yy} (\Delta y)^2 \\
\end{bmatrix}

\quad (A-6)
\]

and inverting the above relation gives
Cubic Form - Coefficients in the cubic expansion correspond to those in a four-point forward difference formula. Because the expansion terms are no higher than third order, the unique coefficient types can be derived by writing the function in terms of only three independent variables, say $x$, $y$ and $z$:

\[ f = f^1 + f_x \Delta x + f_y \Delta y + f_z \Delta z + \frac{1}{2} f_{xx} (\Delta x)^2 + \ldots + \frac{1}{6} f_{xxx} (\Delta x)^3 + \ldots + \frac{1}{6} f_{zzz} (\Delta z)^3 \]  

(A-8)
Referring to Figure A-3, we write

\[
\begin{align*}
\begin{pmatrix}
 f^1 \\
 f^2 \\
 f^3 \\
 f^4 \\
 f^5 \\
 f^6 \\
 f^7 \\
 f^8 \\
 f^9 \\
 f^{10} \\
 f^{11} \\
 f^{12} \\
 f^{13} \\
 f^{14} \\
 f^{15} \\
 f^{16} \\
 f^{17} \\
 f^{18} \\
 f^{19} \\
 f^{20}
\end{pmatrix} &= 
\begin{pmatrix}
 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 0 & 0 & 0 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 \\
 1 & 0 & 1 & 0 & 0 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 \\
 1 & 0 & 0 & 1 & 0 & 0 & 0 & 0 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 \\
 1 & 2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 1 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 0 & 2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 1 & 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 0 & 0 & 2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 3 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 2 & 1 & 0 & 2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 1 & 2 & 0 & 1 & 2 & 2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 0 & 3 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 2 & 0 & 1 & 2 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 & 1 \\
 1 & 0 & 2 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
 1 & 1 & 0 & 2 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \end{pmatrix}
\end{align*}
\]

\[
\begin{align*}
\{ f^1 \} &= f^1_x \Delta x \\
\{ f^2 \} &= f^1_y \Delta y \\
\{ f^3 \} &= f^1_z \Delta z \\
\{ f^4 \} &= f_{xx} (\Delta x)^2 \\
\{ f^5 \} &= f_{xy} \Delta y \Delta x \\
\{ f^6 \} &= f_{yy} (\Delta y)^2 \\
\{ f^7 \} &= f_{zz} \Delta z \Delta y \\
\{ f^8 \} &= f_{z}(\Delta z)^2 \\
\{ f^9 \} &= f_{xxx} (\Delta x)^3 \\
\{ f^{10} \} &= f_{xxy} \Delta y (\Delta x)^2 \\
\{ f^{11} \} &= f_{yy} \Delta x \Delta y \\
\{ f^{12} \} &= f_{yyy} (\Delta y)^3 \\
\{ f^{13} \} &= f_{zxx} \Delta z (\Delta x)^2 \\
\{ f^{14} \} &= f_{zyx} \Delta y \Delta x \\
\{ f^{15} \} &= f_{azy} \Delta y \Delta z \\
\{ f^{16} \} &= f_{zzx} (\Delta z)^2 \Delta x \\
\{ f^{17} \} &= f_{zy} (\Delta z)^2 \Delta y \\
\{ f^{18} \} &= f_{zz} (\Delta z)^3
\end{align*}
\]
The inverse of this matrix is the matrix of difference coefficients, and is given below.

\[
\begin{bmatrix}
1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
-\frac{11}{6} & 3 & 0 & 0 & -\frac{3}{2} & 0 & 0 & 0 & 0 & 0 & \frac{1}{3} & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
-\frac{11}{6} & 0 & 3 & 0 & 0 & -\frac{3}{2} & 0 & 0 & 0 & 0 & 0 & \frac{1}{3} & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
-\frac{11}{6} & 0 & 0 & 3 & 0 & 0 & 0 & 0 & \frac{3}{2} & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & \frac{1}{3} \\
2 & -5 & 0 & 0 & 4 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
2 & -\frac{5}{2} & -\frac{5}{2} & 0 & 1 & 2 & 3 & \frac{1}{2} & 0 & 0 & 0 & 0 & \frac{1}{2} & -\frac{1}{2} & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
2 & 0 & -5 & 0 & 0 & 0 & 4 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
2 & -\frac{5}{2} & 0 & -\frac{5}{2} & 0 & 0 & 3 & 0 & \frac{1}{2} & 0 & 0 & 0 & 0 & -\frac{1}{2} & 0 & 0 & \frac{1}{2} & 0 & 0 & 0 & 0 & 0 & 0 \\
2 & 0 & -\frac{5}{2} & -\frac{5}{2} & 0 & 0 & 1 & 2 & \frac{1}{2} & 0 & 0 & 0 & 0 & 0 & 0 & \frac{1}{2} & 0 & -\frac{1}{2} & 0 & 0 & 0 & 0 & 0 \\
2 & 0 & 0 & -5 & 0 & 0 & 0 & 0 & 0 & 4 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & -1 \\
-1 & 3 & 0 & 0 & -3 & 0 & 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
-1 & 2 & 1 & 0 & -1 & -2 & 0 & 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
-1 & 1 & 2 & 0 & 0 & -2 & -1 & 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
-1 & 0 & 3 & 0 & 0 & 0 & -3 & 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
-1 & 2 & 2 & 1 & -1 & 0 & 0 & -2 & 0 & 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\
-1 & 1 & 1 & 1 & 0 & -1 & 0 & -1 & -1 & 0 & 0 & 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 \\
-1 & 0 & 2 & 1 & 0 & 0 & -1 & 0 & -2 & 0 & 0 & 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 \\
-1 & 1 & 0 & 2 & 0 & 0 & 0 & -2 & 0 & -1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 & 0 \\
-1 & 0 & 1 & 2 & 0 & 0 & 0 & 0 & -2 & -1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 \\
-1 & 0 & 0 & 3 & 0 & 0 & 0 & 0 & -3 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 & 1 \\
\end{bmatrix}
\]
Organization of Terms - The number of unique terms in a symmetric tensor of order $M$ and dimension $N$, is summarized as follows.

<table>
<thead>
<tr>
<th>Order</th>
<th>Number of Terms</th>
</tr>
</thead>
<tbody>
<tr>
<td>0</td>
<td>1</td>
</tr>
<tr>
<td>1</td>
<td>$N$</td>
</tr>
<tr>
<td>2</td>
<td>$N(N+1)/2$</td>
</tr>
<tr>
<td>3</td>
<td>$N(N+1)(N+2)/6$</td>
</tr>
<tr>
<td>$M$</td>
<td>$N(N+1)(N+2) \cdots (N+M-1)/M!$</td>
</tr>
</tbody>
</table>

These relations also give the number of unique $M$th order partial derivatives involved in an $M$th order Taylor series expansion of $N$ variables. For example, the coefficients of a second-order expansion in three variables $(x,y,z)$ are the $3(3+1)/2 = 6$ second partial derivatives $(xx, yx, yy, zx, zy, zz)$. The total number of terms required in the expansion is the sum of the numbers of partial derivatives of each order, for example a second-order expansion in three variables requires $1 + 3 + 3(3+1)/2 = 10$ total terms, and therefore a minimum of $10$ function evaluations to determine the coefficient (partial derivative) values.

The required coefficients for an expansion are conveniently organized into a one dimensional array, for example for a function of three variables the array is

$$(f_1 f_2 f_3 f_{11} f_{21} f_{22} f_{31} \cdots f_{111} \cdots)$$

The terms are then easily retrieved from this array using the tensor relations given above. Thus using $L_i, L_{ij}, L_{ijk}, \ldots$ to denote the location of the respective $i$, $ij$, $ijk$, $\ldots$ derivative terms within the array, we have for a function of $N$ variables

$$
L_0 = 1 \\
L_i = L_0 + i \\
L_{00} = 1 + N \\
L_{ij} = L_{00} + (i-1)(i)/2 + j \\
L_{000} = L_{00} + N(N+1)/2 \\
L_{ijk} = L_{000} + (i-1)(i)(i+1)/6 + (j-1)(j)/2 + k
$$
General Discussion - It may be observed that the forward difference scheme outlined here requires a minimum number of function evaluations, i.e. one plus the number of partial derivatives involved in the expansion. Along with its other advantages of possible improved accuracy in certain types of situations, this would seem to indicate that the forward difference formulas presented here constitute the best approach for the required function expansions.

An interesting alternative derivation of the difference coefficient matrices is possible by a procedure used in the finite element method. The function to be expanded may be thought of as the quantity (say displacement) being interpolated within the finite element. The function is then defined in terms of its values at the nodes (i.e. the independent variable values) times the corresponding shape functions. The required partial derivatives can then be evaluated explicitly at the "origin", i.e. a corner node at which the sides of the element form an orthogonal coordinate system, simply by differentiating the element shape functions. Of course the appropriate finite element shape functions must be available, but if they are then this process allows derivation of the difference formulas without inversion of a matrix. These considerations were initially responsible for the selection of the forward difference approach, and the second and third order coefficients were evaluated in this manner, for example the third order coefficients using the TET 20 element of Argyris et al. (1968).
APPENDIX B: NONCONSERVATIVE LOADING EFFECTS

General Considerations - Several types of nonconservative loading can occur in finite element analysis, for example where the applied generalized nodal forces depend on the system displacements, velocities, or other displacement or deformation parameters. The nonconservative effects considered in the present work are those due to nodal forces which are a function of the nodal displacements, and in particular the effects of pressure loadings where the pressurized surface undergoes significant changes in area and orientation. General formulations may be developed for these cases, in terms of area integrals and pressure intensities. However, in order to illustrate completely the basic effects, the nonconservative load terms are derived here for the special case of a constant-strain-triangle (CST) finite element.

Figure B-1: CST Element for Nonconservative Loading
CST Element Definitions - Referring to Figure B-1, consider the CST element with nodes 1, 2 and 3, and subjected to a normal pressure loading of intensity h. The vectors \( \mathbf{V}_{21} \) and \( \mathbf{V}_{31} \) connect the nodes 1-2 and 1-3, respectively. The cross product \( \mathbf{V}_{21} \times \mathbf{V}_{31} \) is then a vector with (positive) direction normal to the element surface and with magnitude equal to twice the element area.

Coordinates in the initial configuration are defined by the components \( x,y,z \), and corresponding displacements by \( u,v,w \). Then by defining for node \( k \), the quantities

\[
\begin{align*}
\mathbf{c}_x^k &= (x_k + u_k) - (x_1 + u_1) \\
\mathbf{c}_y^k &= (y_k + v_k) - (y_1 + v_1) \\
\mathbf{c}_z^k &= (z_k + w_k) - (z_1 + w_1)
\end{align*}
\]  

we may write the vectors \( \mathbf{V} \) as

\[
\begin{align*}
\mathbf{V}_{21} &= (\mathbf{c}_x^2, \mathbf{c}_y^2, \mathbf{c}_z^2) \\
\mathbf{V}_{31} &= (\mathbf{c}_x^3, \mathbf{c}_y^3, \mathbf{c}_z^3)
\end{align*}
\]  

while their rates are given by

\[
\begin{align*}
\dot{\mathbf{V}}_{21} &= (\dot{u}_2 - \dot{u}_1, \dot{v}_2 - \dot{v}_1, \dot{w}_2 - \dot{w}_1) \\
\dot{\mathbf{V}}_{31} &= (\dot{u}_3 - \dot{u}_1, \dot{v}_3 - \dot{v}_1, \dot{w}_3 - \dot{w}_1)
\end{align*}
\]  

B.1 Fundamental Equilibrium Formulation

In this section the nonconservative loading effects are formulated for the CST element, and used to generalize the fundamental equilibrium equations in Section 2 to analysis of nonconservative systems.
Basic Load Equation - For a uniform pressure loading, each node of the CST element has an equivalent concentrated load vector \( p_i \), given by

\[
p_i = \frac{1}{6} h (v_{21} \times v_{31})_i
\]  

(B-2a)

The nodal force during an increment of loading is defined by the basic nonconservative load equation

\[
p_i = \frac{1}{6} (h^* + \Delta h^*) (v_{21} \times v_{31})_i
\]  

(B-2b)

where \( h^* \) is the pressure intensity at the reference equilibrium configuration, \( h^* \) is a constant pressure distribution and \( \Delta \) is the incremental load parameter.

First-Order Load Equation - Differentiating equation (B-2b) with respect to the fundamental path parameter, gives

\[
\dot{p_i} = \frac{1}{6} \dot{\Lambda} h^* (v_{21} \times v_{31})_i + \frac{1}{6} (h^* + \Delta h^*) (\dot{v}_{21} \times v_{31} + v_{21} \times \dot{v}_{31})_i
\]  

(B-3a)

Evaluating at the reference equilibrium configuration \( (\Lambda = 0, v = v^*, etc.) \) gives

\[
\dot{p_i}^* = \frac{1}{6} \dot{\Lambda}^* h^* (v_{21}^* \times v_{31}^*)_i + \frac{1}{6} h^* (v_{21}^* \times \dot{v}_{31}^* + v_{21}^* \times v_{31}^*)_i
\]  

(B-3b)

This is the first order nonconservative load equation. To put it into the desired form, we write the vector \( q \) of element displacements as

\[
q = (u_1, v_1, w_1, u_2, v_2, w_2, u_3, v_3, w_3)
\]

from which it follows using equations (B-1), that

\[
\frac{1}{6} h (v_{21} \times \dot{v}_{31} + \dot{v}_{21} \times v_{31})_i = C_{ij} \dot{q}_j
\]  

B-3
The desired form of the first order nonconservative load equation is then

\[ \mathbf{\ddot{p}}_i = \mathbf{\ddot{\lambda}} p_i + C_{ij} \mathbf{\ddot{q}}_j \]

where

\[ C_{ij} = \frac{1}{6} h \begin{bmatrix} 0 & cz_2 & -cy_2 & 0 & 0 & -cz_2 & cy_2 \\ -cz_3 & 0 & -cz_3 & cz_3 & -cy_3 & 0 & cy_2 \\ +cz_3 & cx_2 & 0 & -cx_2 & -cx_3 & cx_3 & 0 \\ cy_2 & -cx_2 & 0 & cy_3 & -cx_3 & 0 & -cy_2 \\ -cy_3 & cx_3 & 0 & cx_3 & 0 & 0 & 0 \\ \end{bmatrix} \]

The first part of \( \mathbf{\ddot{p}} \) in (B-3c) is a usual nodal load rate, and occurs in the nonconservative form of equilibrium equation (2-7c) as a contribution to the load term \( P^o \) (see (2-9)). The second part of \( \mathbf{\ddot{p}} \) occurs in the nonconservative form of (2-7c) as an unsymmetric contribution to the Jacobian stiffness \( K^o \) (note that since \( C_{ij} \) occurs on the left-hand-side of (2-7c), it must be subtracted from \( K^o \)).

Second-Order Load Equation - A second differentiation of (B-2b) gives

\[ \mathbf{\dddot{p}}_i = \frac{1}{6} h \mathbf{\dddot{\lambda}} h^o (V_{21} \times V_{31})_i + \frac{1}{3} h \mathbf{\dddot{\lambda}} h^o (V_{21} \times V_{31} + V_{21} \times V_{31})_i \\
+ \frac{1}{6} (h^o + h^o) (V_{21} \times V_{31} + V_{21} \times V_{31} + 2V_{21} \times V_{31}) \]
Evaluating at the reference equilibrium configuration provides

\[ p_i^* = \frac{1}{6} \dot{A}^* \dot{h}^* (\dot{V}_{21}^* x \dot{V}_{31}^*)_i + \frac{1}{3} \ddot{A}^* \dot{h}^* (\dot{V}_{21}^* x \dot{V}_{31}^* + \dot{V}_{21}^* x \dot{V}_{31}^*)_i \]

\[ + \frac{1}{6} \dot{h}^* (\ddot{V}_{21}^* x \dot{V}_{31}^* + \ddot{V}_{31}^* x \dot{V}_{31}^* + 2\dot{V}_{21}^* x \dot{V}_{31}^*)_i \]  

(B-4b)

This is the second order nonconservative load equation, which may be written in the form

\[ p_i^* = C_{ij}^* q_j + p_{li}^* \]  

(B-4c)

where

\[ p_{li}^* = \frac{1}{6} \dot{A}^* \dot{h}^* (\dot{V}_{21}^* x \dot{V}_{31}^*)_i + \frac{1}{3} \ddot{A}^* \dot{h}^* (\dot{V}_{21}^* x \dot{V}_{31}^*)_i \]

\[ + \dot{V}_{21}^* x \dot{V}_{31}^*)_i \]  

(B-5a)

B.2 Bifurcation and Postbuckling Formulation

**Basic Load Equation** - Development of the nonconservative load effects for bifurcation and postbuckling follows the fundamental relations of section B.1. Using equation (B-2b), we may write the nodal force during an increment of loading on the postbuckling path, as

\[ p_i = p_i^f + p_i^p = \frac{1}{6} (h^* + \dot{A} \dot{h}^*) ((V_{21}^f + V_{21}^P) x (V_{31}^f + V_{31}^P))_i \]  

(B-5a)

Since this relation will be used to establish a nonconservative form of the postbuckling equilibrium equation (3-3), the fundamental load contribution must be subtracted from (B-5a). We then obtain

\[ p_i^p = \frac{1}{6} (h^* + \dot{A} \dot{h}^*) (V_{21}^f x V_{31}^P + V_{21}^P x V_{31}^f + V_{21}^P x V_{31}^P)_i \]  

(B-5b)
First-Order Load Equation - Differentiating equation (B-5b) with respect to the postbuckling path parameter, gives

\[ p'_{i} = \frac{1}{6} \Lambda' h^{o} (v_{21}^{f} x v_{31}^{p} + v_{21}^{p} x v_{31}^{f} + v_{21}^{p} x v_{31}^{f})_{i} \]

\[ + \frac{1}{6} (h^{*} + \Lambda h^{o}) (v_{21}^{f} x v_{31}^{p} + v_{21}^{p} x v_{31}^{f} + v_{21}^{p} x v_{31}^{f})_{i} \]

\[ + v_{21}^{p} x v_{31}^{p} + v_{21}^{p} x v_{31}^{f})_{i} \]

(E-6a)

Evaluating at the critical bifurcation point \((V^{p} = 0)\), gives

\[ p'_{i} = \frac{1}{6} (h^{*} + \Lambda h^{o}) (v_{21}^{f} x v_{31}^{p} + v_{21}^{p} x v_{31}^{f})_{i} \]

(B-6b)

Substituting for \(v_{f}^{*}\) gives

\[ p'_{i} = \frac{1}{6} (h^{*} + \Lambda h^{o}) (v_{21}^{f} x v_{31}^{p} + \Delta v_{21}^{f} x v_{31}^{p} + v_{21}^{p} x v_{31}^{f} + v_{21}^{p} x \Delta v_{31}^{f})_{i} \]

(B-6c)

and using the relations which express \(v_{f}^{*}\) in terms of \(q_{p}^{*}\), gives

\[ p'_{i} = C_{ij} q_{j}^{*} + \frac{1}{6} h^{*} (\Delta v_{21}^{f} x v_{31}^{p} + v_{21}^{p} x \Delta v_{31}^{f})_{i} \]

\[ + \frac{1}{6} \Lambda h^{o} (v_{21}^{f} x v_{31}^{p} + v_{21}^{p} x v_{31}^{f} + v_{21}^{p} x v_{31}^{f} + v_{21}^{p} x \Delta v_{31}^{f})_{i} \]

(B-6d)

This is the first order postbuckling nonconservative load expression, which may be written in the form

\[ p'_{i} = C_{ij} q_{j}^{*} + p_{i}^{1} \]

(B-6e)

where

\[ p_{i}^{1} = \frac{1}{6} \Lambda h^{o} (v_{21}^{f} x v_{31}^{p} + v_{21}^{p} x v_{31}^{f})_{i} \]

\[ + \frac{1}{6} (h^{*} + \Lambda h^{o}) (\Delta v_{21}^{f} x v_{31}^{p} + v_{21}^{p} x \Delta v_{31}^{f})_{i} \]

B-6
Expression (B-6e) provides the necessary nonconservative addition to the load term, in the first order postbuckling equation (3-5d). The Jacobian $K_0^*$ again becomes nonsymmetric due to the subtraction of the $C^*$ terms. Thus a nonlinear, nonconservative eigen equation is produced, which may be solved by the same approaches discussed in Section 3 for the symmetric problem.

**Second-Order Load Equation** - A second differentiation of (B-5b) with respect to the postbuckling path parameter, gives

$$p''_i^P = \frac{1}{6} \Lambda 'h^o \left( \nu_{21}' x v_{31}^P + \nu_{21}^P x v_{31}^f + \nu_{31}^P x v_{31}^P \right) i$$

$$+ \frac{1}{3} \Lambda 'h^o \left( \nu_{21}' x v_{31}^P + \nu_{21}^P x v_{31}^f + \nu_{31}^P x v_{31}^f \right) i$$

$$+ \nu_{21}^P x v_{31}^P + \nu_{21}^P x v_{31}^f \right) i$$

$$+ \frac{1}{6} (h^* + \Lambda h^o) \left( \nu_{21}' x v_{31}^P + 2 \nu_{21}^P x v_{31}^P + \nu_{21}^P x v_{31}^f + \nu_{21}^P x v_{31}^f \right) i$$

$$+ 2 \nu_{21}^P x v_{31}^f + \nu_{21}^P x v_{31}^f + \nu_{21}^P x v_{31}^f + \nu_{21}^P x v_{31}^f + \nu_{21}^P x v_{31}^f \right) i$$

Evaluating at the critical point ($v_P^P = 0$), with the critical value of $h = h^* + \Lambda h^o$, gives

$$p''_i^P = \frac{1}{3} \Lambda 'h^o \left( \nu_{21}' x v_{31}^P + \nu_{21}^P x v_{31}^f \right) i$$

$$+ \frac{1}{6} h \left( 2 \nu_{21}' x v_{31}^P + \nu_{21}^P x v_{31}^P + \nu_{21}^P x v_{31}^P + \nu_{21}^P x v_{31}^f \right) i$$

$$+ 2 \nu_{21}^P x v_{31}^f \right) i$$

(B-7a)

Using the relations which express $v''_{P}^P$ in terms of $q''_{P}$, gives

$$p''_i^P = \frac{1}{3} \Lambda 'h^o \left( \nu_{21}' x v_{31}^P + \nu_{21}^P x v_{21}^f \right) i$$

$$+ C_{ij} q''_{i}^P + \frac{1}{6} \left( 2 \nu_{21}' x v_{31}^P + 2 \nu_{21}^P x v_{31}^f + 2 \nu_{21}^P x v_{31}^f \right) i$$

(B-7b)

(B-7c)
This is the second order postbuckling nonconservative load expression, which may be written in the form

\[ p_1''P = C_{ij} q_j'^2 P + 2S_i P_{21}^1 + p_{21}^2 \]  

(B-7d)

where

\[ p_{21}^1 = \frac{1}{6} h \sum_{ij} \left( \frac{\partial^2}{\partial x^2} \right)^i \left( \frac{\partial^2}{\partial y^2} \right)^j \left( \frac{\partial^2}{\partial z^2} \right)^k \left( \frac{\partial^2}{\partial w^2} \right)^l \]  

and

\[ p_{21}^2 = \frac{1}{3} h \sum_{ij} \left( \frac{\partial^2}{\partial x^2} \right)^i \left( \frac{\partial^2}{\partial y^2} \right)^j \left( \frac{\partial^2}{\partial z^2} \right)^k \left( \frac{\partial^2}{\partial w^2} \right)^l \]  

Third-Order Load Equation - A third differentiation of (B-5b) and evaluation at the critical point, provides

\[ p_1'''P = \frac{1}{2} \sum_{ij} \left( \frac{\partial^3}{\partial x^3} \right)^i \left( \frac{\partial^3}{\partial y^3} \right)^j \left( \frac{\partial^3}{\partial z^3} \right)^k \left( \frac{\partial^3}{\partial w^3} \right)^l \]  

and

\[ p_{21}'''P = \frac{1}{6} h \sum_{ij} \left( \frac{\partial^3}{\partial x^3} \right)^i \left( \frac{\partial^3}{\partial y^3} \right)^j \left( \frac{\partial^3}{\partial z^3} \right)^k \left( \frac{\partial^3}{\partial w^3} \right)^l \]  

(B-8a)

B-8
Using the relations which express \( V''''P \) in terms of \( q''''P \), gives

\[
p''''P = \frac{1}{2} \Lambda''' h^o \left( V_{21}^P xV_{31}^P + V_{21}^P xV_{31}^f \right) i + \frac{1}{2} \Lambda''' h^o \left( V_{21}^f xV_{31}^P \right) i + 2V_{21}^f xV_{31}^P \]

\[
+ \frac{1}{2} h \left( V_{21}^f xV_{31}^P + V_{21}^P xV_{31}^f + 2V_{21}^f xV_{31}^P \right) i + C_{ij} q''''P_i
\]

\[
+ \frac{1}{2} h \left( V_{21}^f xV_{31}^P + V_{21}^P xV_{31}^f + V_{21}^P xV_{31}^f + V_{21}^P xV_{31}^P \right) i
\]

This is the third order postbuckling nonconservative load expression, which may be written in the form

\[
p''''P = C_{ij} q''''P_i + 3 \left( s''P_{21} + p_{31} \right)
\]

where

\[
p_{31} = s'' \left( \frac{1}{6} h^o \left( V_{21}^f xV_{31}^P + V_{21}^P xV_{31}^f \right) i + \frac{1}{3} \Lambda''' h^o \left( V_{21}^f xV_{31}^P + V_{21}^P xV_{31}^f \right) i \right.
\]

\[
+ \frac{1}{6} h \left( V_{21}^f xV_{31}^P + V_{21}^P xV_{31}^f \right) i \}
\]

\[
+ s' \left( \frac{1}{6} \Lambda''' h^o \left( V_{21}^f xV_{31}^j + V_{21}^P xV_{31}^b + 2V_{21}^P xV_{31}^j \right) i \right.
\]

\[
+ \frac{1}{6} h \left( V_{21}^f xV_{31}^j + \frac{1}{6} h \left( V_{21}^f xV_{31}^j + V_{21}^P xV_{31}^b \right) i \right)
\]

\[
+ \frac{1}{6} h \left( V_{21}^f xV_{31}^j + V_{21}^P xV_{31}^j \right) i
\]
This appendix contains a FORTRAN IV listing of the PANES (Program for Analysis of Nonlinear Equilibrium and Stability) program. Following the program listing is a listing of input data for the torus problem described in section 5.3.
**P A N E S (PROGRAM FOR ANALYSIS OF NONLINEAR EQUILIBRIUM/STABILITY)**

R.G. VOS, THE BOEING COMPANY, PHONE 773-2638, KENT, WASHINGTON

PANES IBM 360 VERSION (78 DOF) DATED 09/30/74

---

**APPPLICABLE TO NONLINEAR NONCONSERVATIVE HYPERELASTIC SYSTEMS.**

CST ELEMENT, 26 NODES, 78 DOF, 24 ELEMENTS, 20 LOAD INCREMENTS.

5 MATERIALS, 2 OR 3 DOF PER NODE.

NODES ARE LOCATED IN BASIC-CARTESIAN OR CYLINDRICAL COORDINATES

AND DISPLACEMENTS ARE IN BASIC, CYLINDRICAL, OR SPECIAL-CARTESIAN.

INTEGER UIN1, UIN2, UOUT, UINRS, UOUTRS

INTEGER NOD, NEL, NN, NF, NS, ORD, I, MJUMP, NJUMP

INTEGER IPRESS, PREU, IPRED(20), MAXUP, MAXIT, NMAT, IMAT(24)

INTEGER NINCR, INCR, IPRED, ITER, ELNO(3, 24), KFD(78), IDET

DOUBLE PRECISION PFACT(2, 20), PREF(2, 78), P0(2), P1(2), PA(2)

DOUBLE PRECISION PRFACT(20), PRREF(24), PRO, PRI, PRA

DOUBLE PRECISION P(78), Q(78), R(78), RR(78), PUUM(78)

DOUBLE PRECISION LSIGN, RL, RRL, PATH, ERR, ERRMAX, DET

DOUBLE PRECISION LAM, LAMR, LAM, FLAMAX, LAMIN, JUMP, JRUMP, SLOPED

DOUBLE PRECISION COORD(5), E(5), NU(5), C, DFE, DFF

DOUBLE PRECISION GCOS(9, 26), COORD(3, 26), KMAT(78, 78)

DOUBLE PRECISION T(24), GEOM(3, 24), EET(3, 24)

COMMON/COMNEL/NEL/COMMON/COMNS/NS/COMMON/ORD/ORD/COMMON/DFE/DFE/COMMON/EET

COMMON/COMNF/NF/COMMON/IMAT/IMAT/COMMON/EU/EU/COMMON/COMC/Q

COMMON/COMCOS/GCOS/COMMON/COORD/COORD/COMMON/ELNO/COMMON/E/COMMON/E/COMMON/T

COMMON/COMPR/IPRESS/COMMON/COMPR/PR/COMMON/PR/PRREF

---

**START PROBLEM**

1 CALL READRS(UIN1, UIN2, UOUT, INCR, UINRS, UOUTRS)

CALL READ0(I, UOUT, ORD, PREU, MAXUP, MAXIT, ERRMAX, DFE, DFF, MJUMP, JUMP, SLOPED, FLAMAX, LAMIN)

IF(UINRS .GT. 0) GO TO 11

COLD START

CALL BIGS(1, UIN1, UOUT, IPRESS, NF, NMAT, E, NU, NORDA, NOD, NEL, COORD, GCOS, IMAT, T, ELNO, GEOM, KFD, PREF, PRREF, LSGN, PP, QQ, P1, PRI)

NJUMP = MJUMP + 1

GO TO 16

READ RESTART TAPE

11 CALL BIGS(2, UINRS, INCR, IPRESS, NF, NMAT, E, NU, NORDA, NOD, NEL, COORD, GCOS, IMAT, T, ELNO, GEOM, KFD, PREF, PRREF)
C GENERAL PROGRAM FLOW

17 NS = 3
NN = NOD*NF
C READ INCREMENTAL LOAD FACTOR DATA
CALL REACI(5,UOUT,NINCR,PRED,IPRED,PFACT,PRFACT)
C BEGIN LOAD INCREMENT LOOP
DO 1000 INCR=1,NINCR
C LAM = INCREMENTAL LOAD PARAMETER (MAXIMUM VALUE 1.0).
C LAMR = LOAD YET TO BE APPLIED = 1.0 - LAM.
C LAMS = LOAD STEP PARAMETER = FRACTION OF LAMR TO BE APPLIED.
LAM = 0.00
C LSIGN = +,- FOR LOADING,UNLOADING SITUATION.
IF(LSIGN.LT.0.0)STOP
C BEGIN LOAD STEP
ISTEP = ISTEP+1
NUP = 0
LAMR = 1.00-LAM
C SET UPPER BOUND FOR ABSOLUTE VALUE OF LOAD STEP SIZE LAMS.
LAMS = 1.00
IF(LSIGN.LT.0.0)LAMS = FLAMAX/LAMR
C CALL PFORCE TO GIVE APPLIED CONSERVATIVE NODAL LOADS P.
CALL EFORCE TO GIVE APPLIED NONCONSERVATIVE NODAL LOADS Q.
CALL PFORCE(P1,PREF,NN,P)
CALL EFORCE(IPRESS,PR1,PRREF,QQ,NEL,NN,NF,ELNO,Q)

COMPUTE LOAD STEP NODAL LOADS. THESE = APPLIED LOAD - INTERNAL
LOAD FOR SPECIFIED FORCE DOF, APPLIED DISPLACEMENT - CURRENT
DISPLACEMENT FOR SPECIFIED DISPLACEMENT DOF.

DO 210 I=1,NN
  IF(KFD(I).GT.0) P(I) = P(I) + Q(I) - PP(I)
  IF(KFD(I).LT.0) P(I) = P(I) - Q(I)
210

FORM JACOBIAN AT BEGINNING OF LOAD STEP.
CALL MERGE(KMAT,ELNO,KFD,NEL,NN,NF)
CALL DECOMP(KMAT,NN,KFD,DET,DET)

FUNDAMENTAL PATH PREDICTOR CODE

IF(IPRED(INCR).GE.2)GO TO 241

APPLY LINEAR PREDICTOR FOR LOAD STEP
CALL SOLVE(KMAT,NN,KFD,P,Q)
LAMS = LSIGN*LAMS
DO 230 I=1,NN
  Q(I) = LAMS*Q(I)
230 GC

APPLY QUADRATIC PREDICTOR FOR LOAD STEP

241 CALL RATES(KMAT,POUM,NN,KFD,P,PRA,PR1,LSIGN,RL,RL,RRQ)
  RJUMP = JUMPP/LAMR
  CALL STEP(LSIGN,RL,RL,NN,RL,RRQ,RJUMP,JUMPP,NJUMP,SLOPED, 1PATH,LAMS)
  IF LIMIT POINT WAS TRAVERSED (I.E. LAMS = 0) OUTPUT LIMIT RESULTS.
  IF(LAMS.EQ.0.DO)
1CALL OUTLIM(6,NOD,NEL,NN,NF,ELNO,EDEOM,EET,CC,P,Q,DOFF, 2RL,RL,RRQ,RRQ,LAMR,LMR)

THIS SECTION OF CODE IS TEMPORARY POSTBuckLING CHECKOUT CODE.

********** THIS SECTION OF CODE IS TEMPORARY POSTBuckLING CHECKOUT CODE.**

** INTEGER BCODE,PCODE,IPOST
** DOUBLE PRECISION SCRIT,LCRIT,LDOT1,LDOT2,QQDOT1(78),QQDOT2(78), 1LPOST1,LPOST2,QQPOS1(78),QQPOS2(78),QQPOS3(78),QQCRIT(78)
** DOUBLE PRECISION POUM,POUM2,POUM3,POUM4(78)

SET BCODE=1 TO GET NONLINEAR EIGEN SOLUTION FOR BIFURCATION POINT.
ALSO SET PCODE=1 TO GET ADDITIONAL PBSTBuckLING PATH SOLUTION.

BCODE = 0
PCODE = 1
IF (PCODE.EQ.0) GO TO 279
CALL EIGEN(UOUT, KMAT, PDUM, Q, NN, KFD, PRA, PR1, RL, RRL, RG, RRQ, 25, 15, 1.0-2, 1.0-5, SCRIT, QQPOS1, IPOST)
CALL OUTPQ(UOUT, NOQ, NF, QQPOS1, QQPOS1)
LCRIT = RL*SCRIT + 5.0D0*RRL*SCRIT**2
IF (PCODE.EQ.0) GO TO 279
PERFORM POSTBUCKLING SOLUTION
LCRIT = RL*SCRIT
LDOT1 = RL
LDOT2 = 0.00
DO 265 I = 1, NN
QQCRIT(I) = QQ(I) + RQ(I)*SCRIT
QQDOT1(I) = RQ(I)
265 QQDOT2(I) = 0.00
IF (IPRED(INCR).LT.2) GO TO 271
LCRIT = LCRIT + 5.0D0*RRL*SCRIT**2
LDOT1 = LDOT1 + RRL*SCRIT
LDOT2 = RRL
DO 270 I = 1, NN
QQCRIT(I) = QQCRIT(I) + 5.0D0*RQ(I)*SCRIT**2
QQDOT1(I) = QQDOT1(I) + RQ(I)*SCRIT
270 QQDOT2(I) = RQ(I)
CALL OUTPQ(UOUT, NOQ, NF, QQDOT1, QQDOT2)
DO 268 I = 1, NN
C = QQ(I)
QQ(I) = QQCRIT(I)
268 QQCRIT(I) = C
271 CALL MERGE(KMAT, ELNO, KFD, NEL, NN, NF)
DO 272 I = 1, NN
C = QQ(I)
QQ(I) = QQCRIT(I)
272 QQCRIT(I) = C
IPOST = KFD(IPOST)
KFD(IPOST) = -IPST
CALL DECOMP(KMAT, NN, KFD, IDET, DET)
CALL PRATES(KMAT, PDUM, PDUM2, PDUM3, PDUM4, NN, IPOST, KFD, SCRIT, 1CQCRIT, QQDOT1, QQDOT2, LCRIT, LDOT1, LDOT2, PR0, PRA, 2QQPOS1, LPOST1, QQPOS2, LPOST2, QQPOS3)
CALL OUTPQ(UOUT, NOQ, NF, QQCRIT, QQPOS1)
CALL OUTPQ(UOUT, NOD, NF, QQPOS2, QQPOS3)
KFD(IPOST) = IPOST

C*******************************************************************************
C PATH CONTINUATION CODE
C*******************************************************************************
279 CONTINUE
DC 280 I=1, NN
280 Q(I) = PATH*RQ(I) + .500*PATH**2*RRQ(I)
C ACD STEP LOAD LAMS TO INCREMENT LOAD SUM LAM
LAMS = LAMS*LAMR
301 LAM = LAM + LAMS
DO 305 I=1, 2
305 PA(I) = PO(I) + LAM*(PI(I)-PO(I))
CALL PFORCE(PA, PREF, NN, P)
DC 310 I=1, NN
IF(KFD(I).GT.O) QQ(I) = QQ(I) + Q(I)
IF(KFD(I).LT.O) QQ(I) = P(I)
310 CONTINUE
PRA = PRO + LAM*(PR1-PRO)
ITER = 0
GO TO 451
C BEGIN ITERATION LLCP
311 ITER = ITER+1
IF(ITER.GT.I)GO TO 401
C FORM JACOBIAN AT APPROXIMATE END OF STEP
CALL MERGE(KMAT, ELNO, KFD,NEL, NN, NF)
CALL DECOMP(KMAT, NN, KFD, IDET, DET)
401 CALL SOLVE(KMAT, NN, KFD, PQ)
C UPDATE INTERNAL FORCES PP AND DISPLACEMENTS QQ.
C COMPUTE APPLIED EXTERNAL LOADS (CONSERVATIVE P, NONCONSERVATIVE Q)
DC 410 I=1, NN
410 QQ(I) = QQ(I) + Q(I)
451 CALL FORCE(NEL, NN, NF, ELNO, DFF, QQ, PP)
CALL PFORCE(PA, PREF, NN, P)
CALL EFORCE(IPRESS, PRA, PRREF, QQ, NEL, NN, NF, ELNO, Q)
DC 460 I=1, NN
C = 0.0
IF(KFD(I).GT.O) C = P(I) + Q(I) - PP(I)
460 P(I) = C
CALL ERCOMP(UOUT, NN, KFD, PP, P, ERR)
END ITERATION LOOP

IF(ERR.LE.ERRMAX)GO TO 901
IF(ITER.LT.MAXIT)GO TO 311

IF THIS POINT IS REACHED, MAX. NC. ITERATIONS OCCUR

IF(NUP.GE.MAXUP)GO TO 901
NUP = NUP+1
ITER = 0
GO TO 311

OUTPUT INCREMENTAL STEP RESULTS

901 CALL HEAD(UOUT,INCR,ISTEP,LAMS,LAM,PFAC(1,INCR),PFAC(2,INCR),
IPRFAC(INCR),IPRED(INCR),MAXUP,NUP,MAXIT,ITER,ERRMAX,ERR)

CALL OUTPUT(UOUT,NOD,NF,PP,QQ)
CALL STRAIN(NEL,ELNO,EGEOM,NF,QQ,EET)
CALL OUTE(UOUT,NEL,EET)

END LOAD STEP

IF(LAM.LT.LAMIN)GO TO 1
IF(LAM.LT.99900)GO TO 201
IF(UOUTRS.EQ.0)GO TO 1000

WRITE INCREMENTAL RESTART TAPE

1000 CONTINUE

END LOAD INCREMENT LOOP

END

SUBROUTINE BIGS(KODE,II,12,IPRESS,NF,NMAT,E,NU, 00002300
1CORDA,NOD,NEL,COORD,GCOS,IMAT,T,ELNO,EGEOM,KFD,PREF,PRREF,
2LSIGN,PP,QQ,PI,PR1)

KCODE = 1,2,3 = INITIALIZE, READ RESTART, WRITE RESTART.
II = INPUT OR RESTART INPUT-OUTPUT FILE UNIT NUMBER.
I2 = OUTPUT FILE UNIT NUMBER, OR INCREMENT NUMBER FOR RESTART.
IPRESS = NONCONSERVATIVE CODE = 0,1 FOR NO PRESSURE,PRESSURE.
NF = NUMBER OF FREEDOMS PER NODE.
NMAT = NUMBER OF MATERIALS.
E,NU = MATERIAL CONSTANTS (E.G. ELASTIC MODULUS,POISSON'S RATIO).
1CORDA(I) = ANGLE FOR SPECIAL COORDINATE SYSTEM I.
NOD,NEL = NUMBER OF NODES,ELEMENTS.
COORD(J,I) = COORDINATES OF NODE I.
GCOS(J,I) = DIRECTION COSINES FOR NODE I.
IMAT(I) = MATERIAL NUMBER FOR ELEMENT I.
T(I) = THICKNESS OF ELEMENT I.
ELNO(J,I) = NODE NUMBERS FOR ELEMENT I.
EGEOM(J,I) = GEOMETRY FOR ELEMENT I (BASE,HEIGHT,PART BASE).
KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.
PREF(J,I) = NODAL LOAD AT DOF I FOR LOAD REFERENCE VECTOR J.
PRREF(I) = INTENSITY ON ELEMENT I FOR PRESSURE REFERENCE VECTOR.
LSIGN = +/- IF LOAD PARAMETER IS INCREASING, DECREASING.
PP(I) = CURRENT INTERNAL NODAL FORCE AT DOF I.
CQ(I) = CURRENT NODAL DISPLACEMENT AT DOF I.
PI(J) = LOAD FACTOR TO BE APPLIED TO PREF(J,I).
PR1 = LOAD FACTOR TO BE APPLIED TO PRREF(I).
INTEGER KODE,I1,I2,NPRESS,NNM,NNMAT,NNEL,NNMAT(1),NNLNC(3,1),NNKFD(1)
DOUBLE PRECISION E(1),NU(1),COORDA(1),GCCS(9,1),
IIPAT,THICK,T,ELNO,EGEOM
IT(I),EGEOM3(1),PREF(1),PRREF(1),LSIGN,PP(1),CQ(1),PI(I),PR1
EQUIVALENCE (J1,UIN,UINRS,UOUTRS),(J2,UOUT,INCR)
INTEGER J1,UIN,UINRS,UOUTRS,J2,UOUT,INCR
INTEGER INC,J,NN
DOUBLE PRECISION THICK
J1 = 11
J2 = 12
IF(KODE.NE.1)GO TO 101

CALL REAC1(UIN,UOUT,IPRESS,NF,THICK,NMAT,E,NU)
CALL READC(UIN,UOUTCOORDA)
CALL REACM(UIN,UOUT,NOD,NNMAT,E,COORDA,COORD,GCOS,
NIMAT,THICK,T,ELNO,EGEOM)
CALL READK(UIN,UOUT,NOD,NF,KFD)
CALL READP(UIN,UOUT,NOD,NF,KFD,PREF)
[IF KNPRESS.GT.0)CALL READPR(UIN,UOUT,NEL,PRREF)
LSIGN = 1.0
NN = NOD*NF
CC 50 I=1,NN
PP(I) = 0.0
CC 60 I=1,2
50 CQ(I) = 0.0
CC 60 I=1,2
60 PI(I) = 0.0
PR1 = 0.0

RETURN
101 IF(KODE.NE.2)GO TO 201

C READ VARIABLES FROM RESTART TAPE
READ(UINRS)IPRESS,NF,NMAT,(E(I),NU(I),I=1,NMAT),
INCD,NEL,((COORD(J,I),J=1,3),GCOS(J,I),J=1,9),I=1,NOD),
2(IMAT(I),T(I),(ELNO(J,I),EGEOM(J,I),J=1,3),PRREF(I),I=1,NEL)
NN = NOD*NF
READ(UINRS)(KFO(I),(PREF(J,I),J=1,2),I=1,NN)
151 READ(UINRS)INC
IF(INC.EQ.INCR)GO TO 161
READ(UINRS)
GC TO 151
161 READ(UINRS)LSIGN,(PP(I),CQ(I),I=1,NN),(P1(I),I=1,2),PR1
REWIND UINRS
RETURN
201 IF(KODE.NE.3)RETURN

C WRITE VARIABLES ONTO RESTART TAPE
IF(INCR.GT.0)GO TO 251
WRITE(UOUTRS)IPRESS,NF,NMAT,(E(I),NU(I),I=1,NMAT),
INCD,NEL,((COORD(J,I),J=1,3),GCOS(J,I),J=1,9),I=1,NOD),
2(IMAT(I),T(I),(ELNO(J,I),EGEOM(J,I),J=1,3),PRREF(I),I=1,NEL)
NN = NOD*NF
WRITE(UOUTRS)(KFO(I),(PREF(J,I),J=1,2),I=1,NN)
251 WRITE(UOUTRS)INCR
WRITE(UOUTRS)LSIGN,(PP(I),CQ(I),I=1,NN),(P1(I),I=1,2),PR1
RETURN
END
SUBROUTINE READO(UI, UO, ORD, PRED, MAXUP, MAXIT, ERRMAX, DFE, DFF,  
MJUMP, JUMPR, SLOPED, FLAMAX, LAMIN)  
READ PROBLEM IDENTIFICATION AND INCREMENTAL-ITERATIVE CONSTANTS.  
UI, UO = INPUT, OUTPUT FILE UNIT NUMBERS.  
ORD = MAXIMUM TENSOR ORDER TO BE USED FOR STRESS-STRAIN EXPANSION.  
PRED = DEFAULT SOLUTION PREDICTOR ORDER.  
MAXUP = MAXIMUM NUMBER OF JACOBIAN UPDATES PER LOAD STEP.  
MAXIT = MAXIMUM NUMBER OF RESIDUAL-LOAD ITERATIONS PER UPDATE.  
ERRMAX = MAXIMUM ALLOWABLE ERROR NORM.  
DFE, DFF = FINITE DIFFERENCE STEP SIZES TO BE USED IN COMPUTING  
STRESS-STRAIN TENSORS, FORCES.  
MJUMP = NUMBER OF INCREMENT DIVISIONS TO PERFORM WHEN NEARING A  
LIMIT POINT.  
JUMPR = FRACTION OF LOAD INCREMENT PRECEDING LIMIT POINT  
AT WHICH LIMIT IS TO BE TRAVERSLED.  
SLOPED = MAXIMUM SLOPE RATIO (CHANGE/AVERAGE) DURING LOAD STEP.  
FLAMAX = MAXIMUM FRACTION OF LOAD INCREMENT TO BE TAKEN DURING  
NEGATIVE LOADING (AFTER MAXIMUM LIMIT POINT).  
LAMIN = MINIMUM (NEGATIVE) FRACTION OF LOAD INCREMENT AT WHICH  
ANALYSIS IS TERMINATED (AFTER MAXIMUM LIMIT POINT).  
INTEGER UI, UO, ORD, PRED, MAXUP, MAXIT, MJUMP  
DOUBLE PRECISION ERRMAX, DFE, DFF, JUMPR, SLOPED, FLAMAX, LAMIN
INTEGER I,BLANK,IDENT(20)
101 FORMAT(20A4) 00003580
102 FORMAT(4I10,F10.0) 00003590
103 FORMAT(2F10.0) 00003600
104 FORMAT(I10,4F10.0) 00003610
201 FORMAT(1HO,20A4) 00003620
202 FORMAT(/1H,**TENSOR ORDER =**,15/I1H,**PREDICTCR TYPE =**,15/
11H,**MAXIMUM JACOBIAN UPDATES PER STEP =**,15/
21H,**MAXIMUM RESIDUAL LOAD CORRECTIVE ITERATIONS =**,15/
31H,**MAXIMUM ERROR NORM =**,E12.5) 00003630
203 FORMAT(/1H,**DFE =**,E12.5/I1H,**DFF =**,E12.5) 00003640
204 FORMAT(/1H,**MJUMP =**,15/I1H,**JUMPR =**,E12.5/
11H,**SLOPED =**,E12.5/I1H,**FLAMAX =**,E12.5/I1H,**LAMIN =**,E12.5) 00003650
DATA BLANK/' 00003660
READ(UI,101)(IDENT(I),I=1,20) 00003670
WRITE(UO,201)(IDENT(I),I=1,20) 00003680
READ(UI,102)ORD,PRED,MAXUP,MAXIT,ERRMAX 00003690
IF(ORD.EQ.0)ORD = 3 00003700
IF(PRED.EQ.0)PRED = 2 00003710
IF(MAXUP.EQ.0)MAXUP = 0 00003720
IF(MAXIT.EQ.0)MAXIT = 5 00003730
IF(ERRMAX.EQ.0.0)ERRMAX = 1.D-8 00003740
WRITE(UO,202)ORD,PRED,MAXUP,MAXIT,ERRMAX 00003750
READ(UI,103)DFE,DFF 00003760
IF(DFE.EQ.0.0)DFE = 1.D-3 00003770
IF(DFF.EQ.0.0)DFF = 1.D-8 00003780
WRITE(UO,203)DFE,DFF 00003790
READ(UI,104)MJUMP,JUMPR,SLOPED,FLAMAX,LAMIN 00003800
IF(MJUMP.EQ.0)MJUMP = 3 00003810
IF(JUMPR.EQ.0.0)JUMPR = 0.100 00003820
IF(SLOPED.EQ.0.0)SLOPED = 0.500 00003830
IF(FLAMAX.EQ.0.0)FLAMAX = 1.00 00003840
IF(LAMIN.EQ.0.0)LAMIN = 0.00 00003850
WRITE(UO,204)MJUMP,JUMPR,SLOPED,FLAMAX,LAMIN 00003860
RETURN 00003870
END 00003880

SUBROUTINE READL(UI,UC,IPRESS,NF,THICK,NMAT,E,NU) 00003950
READ BASIC CODES AND CONSTANTS.
C UI,UO = INPUT,OUTPUT FILE UNIT NUMBERS.
C IPRESS = NONCONSERVATIVE CODE = 0,1 FOR NO PRESSURE, PRESSURE.
C NF = NUMBER OF FREEDOMS PER NODE.
C THICK = DEFAULT ELEMENT THICKNESS.
C NMAT = NUMBER OF MATERIALS.
C E,NU = MATERIAL CONSTANTS (E.G. ELASTIC MODULUS, POISSON'S RATIO).
 INTEGER UI,UO,IPRESS,NF,NMAT
 DOUBLE PRECISION THICK,E(1),NU(1)
 INTEGER M,N1
 INTEGER, Parameter :: NMAT = 0
 101 FORMAT(2110,F10.0)
 102 FORMAT(2110,F10.0)
 201 FORMAT(1HO,'PRESSURE CODE =',I5/1H,'DOF PER NODE =',I5/
 1/H,'DEFAULT THICKNESS =',E12.5)
 202 FORMAT(/1H,'MATERIAL CONSTANT CONSTANT')
 203 FORMAT(1H,5x,15I5,5x,E12.5,1x,E12.5)
 READ(UI,101)IPRESS,NF,THICK
 IF(NF.EQ.0)NF = 3
 IF(THICK.EQ.0.DO)THICK = 1.0D0
 WRITE(UO,201)IPRESS,NF,THICK
 WRITE(UO,202)
 NMAT = 0
 40 READ(UI,102)I,C1,C2
 IF(I.LE.0)RETURN
 WRITE(UO,203)I,C1,C2
 IF(I.LE.NMAT)GO TO 51
 N1 = NMAT+1
 DO 45 M=N1,I
 E(M) = 0.0D0
 45 NU(M) = 0.0D0
 NMAT = I
 51 E(1) = C1
 NU(1) = C2
 GO TO 40
 END

SUBROUTINE READC(UI,UO,CCORDA)
C READ SPECIAL CARTESIAN COORDINATE SYSTEMS.
C UI,UO = INPUT,OUTPUT FILE UNIT NUMBERS.
CCORDA(I) = ANGLE FOR SPECIAL COORDINATE SYSTEM I.
INTEGER UI, UO
DOUBLE PRECISION COORDA(I)
INTEGER I
DOUBLE PRECISION ANGLE,F
101 FORMAT(I10,F10.0)
201 FORMAT(/1H,'CARTESIAN COORDINATE SYSTEMS DEFINED'/
1H,'NUMBER X-AXIS ANGLE')
202 FORMAT(1H,15,5X,F10.4)
F = 3.141592653589793D0/180.D0
WRITE(UO,201)
DO 5 I=1,5
5 COORDA(I) = 0.D0
READ(UI,101)I,ANGLE
IF(I.LE.0) RETURN
WRITE(UO,202)I,ANGLE
COORDA(I) = ANGLE*F
GO TO 10
END

SUBROUTINE READM(UI, UO, NOD, NEL, COORDA, COORD, GCOS,
IMAT, THICK, T, ELNO, EGEOM)
READ MESH DATA.
UI, UO = INPUT, OUTPUT FILE UNIT NUMBERS.
NOD, NEL = NUMBER OF NOD, ELEMENTS.
COORDA(I) = ANGLE FOR SPECIAL COORDINATE SYSTEM I.
COORD(J, I) = COORDINATES FOR NODE I.
GCOS(J, I) = DIRECTION COSINES FOR NODE I.
IMAT(I) = MATERIAL NUMBER FOR ELEMENT I.
THICK = DEFAULT THICKNESS.
T(I) = THICKNESS OF ELEMENT I.
ELNO(J, I) = NODE NUMBERS FOR ELEMENT I.
EGEOM(J, I) = GEOMETRY FOR ELEMENT I (BASE, HEIGHT, PART BASE).
INTEGER UI, UO, NOD, NEL, IMAT(1), ELNO(3,1)
DOUBLE PRECISION COORDA(1), COORD(3,1), THICK, VLENTH, VDOT
DOUBLE PRECISION GCOS(3,3,1), EGEOM(3,1), T(I)
101 FORMAT(215,3F10.0,15)
102 FORMAT(215,F10.0,315)
201 FORMAT(1H1,"** NODE ***/1H , * NO.  I.D.  LCOCRD  X (R) "
17X,*Y (THETA)  Z (DCOORD*)
202 FORMAT(1H1,215,1X,I5,2X,E12.5,1X,E12.5,1X,E12.5,2X,15)
203 FORMAT(1H1," ELEMENT */
11H , * NO.  I.D.  MATERIAL  THICKNESS* , 4X,
2*NODE 1  NODE 2  NODE 3  AREA*)
204 FORMAT(1H1,215,5X,I5,5X,12.4,3X,15,5X,15,5X,15,5X,E12.4)
F = 3.14159265358979300/180.DO
WRITE(UO,201)
NCO = 0
6 READ(U1,101)I,LCCORD,X,Y,Z,DCOORD
IF(I.LE.0)GO TO 150
NOD = NOD+1
WRITE(UO,202)NOD,I,LCCORD,X,Y,Z,DCOORD
COORD(3,NOD) = Z
IF(LCOORD.EQ.0)GO TO 7
ANGLE = Y*F
Y = X*D(*IN(ANGLE))
X = X*D(*COS(ANGLE))
7 CCORD(1,NOD) = X
COORD(2,NOD) = Y
IF(DCOORD+1)13,12,11
11 ANGLE = COORDA(DCOORD)
GCOS(1,1,NOD) = DCOS(ANGLE)
GCOS(1,2,NOD) = DSIN(ANGLE)
GO TO 100
12 R = DSQRT(X**2 + Y**2)
IF(R.EQ.0.DO)GO TO 13
GCOS(1,1,NOD) = X/R
GCOS(1,2,NOD) = Y/R
GO TO 100
13 GCOS(1,1,NOD) = 1.DO
GCOS(1,2,NOD) = 0.DO
100 GCOS(1,3,NOD) = 0.DO
GCOS(2,1,NOD) = -GCOS(1,2,NOD)
GCOS(2,2,NOD) = GCOS(1,1,NOD)
GCOS(2,3,NOD) = 0.DO
GCOS(3,1,NOD) = 0.DO
GCOS(3,2,NOD) = 0.DO
GCOS(3,3,NOD) = 1.00
GO TO 6
150 WRITE(UO,203)
NEL = 0
151 READ(UI,102)I,M,TT,N1,N2,N3
IF(I.LE.0)GO TO 250
IF(TT.EQ.0.DO)TT = THICK
NEL = NEL+1
IMAT(NEL) = M
T(NEL) = TT
170 J=1,3
V21(J) = CCORD(J,N2) - CCORD(J,N1)
V31(J) = CCORD(J,N3) - CCORD(J,N1)
B = VLENTH(V21)
L31 = VLENTH(V31)
B1 = VDOT(V21,V31)/B
H = DSQRT(L31**2 - B1**2)
A = .500*B*H
EGEOM(1,NEL) = B
EGEOM(2,NEL) = H
EGEOM(3,NEL) = B1
200 WRITE(UO,204)NEL,IM,TT,N1,N2,N3,A
NELIN(NEL) = N1
NELN(2,NEL) = N2
NELN(3,NEL) = N3
GO TO 151
250 RETURN
END

SUBROUTINE READK(UI,UO,NOD,NF,KFD)
C READ SPECIFIED FORCE-DISPLACEMENT-CONSTRAINT DEGREES OF FREEDOM.
C UI,UO = INPUT,OUTPUT FILE UNIT NUMBERS.
C NCD = NUMBER OF NODES.
C NF = NUMBER OF FREEDOMS PER NODE.
C KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.
INTEGER UI,UO,NOD,NF,KFD(1)
INTEGER ISTOR(4),JSTOR(4),KSTOR(4),LSTOR(4)
INTEGER I,J,K,L,N,NN,LLCC,LLCG
101 FORMAT(4(4I5))
201 FORMAT(/11H,'SPECIFIED FORCE-DISPLACEMENT-CONSTRAINT COF' 00005500
1/1H,'NODE I.D. COMPONENT NODE I.D. COMPONENT') 00005510
202 FORMAT(11H,15,6X,15,6X,15,6X,15) 00005520
00005530
WRITE(UO,201) 00005540
C SET DEFAULT CODES TO SPECIFIED FORCE. 00005550
DO 5 I=1,NN 00005560
5 KFD(I) = I 00005570
10 READ(UI,101)(ISTOR(K),JSTOR(K),KSTOR(K),LSTOR(K),K=1,4) 00005580
DO 12 M=1,4 00005590
IF(ISTOR(M).NE.0)GO TO 13 00005590
12 CONTINUE 00005600
GO TO 51 00005610
13 DO 20 M=1,4 00005620
I = ISTOR(M) 00005630
IF(I.LE.0)GO TO 20 00005640
J = JSTOR(M) 00005650
K = KSTOR(M) 00005660
L = LSTOR(M) 00005670
WRITE(UO,202)I,J,K,L 00005680
IF(K.EQ.0)K = I 00005690
IF(L.EQ.0)L = -J 00005700
JLOC = NF*(I-1) + J 00005710
LLOC = L 00005720
IF(L.LT.0)LLOC = -L 00005730
LLOC = NF*(K-1) + LLOC 00005740
IF(L.LT.0)LLOC = -LLOC 00005750
KFD(JLOC) = LLOC 00005760
20 CONTINUE 00005770
GO TO 10 00005780
51 RETURN 00005790
END 00005800

SUBROUTINE READP(UI,UO,NOD,NF,KFD,PREF) 00005810
C READ LOAD REFERENCE CURVES. 00005820
C UI,UO = INPUT,OUTPUT FILE UNIT NUMBERS. 00005830
C NCO = NUMBER OF NODES. 00005840
C NF = NUMBER OF FREEDOMS PER NODE. 00005850
C KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR-DOF I. 00005860
C

PREF(J,I) = NODAL LOAD AT CGF I FOR LOAD REFERENCE VECTOR J.

INTEGER UI, UO, NO, NF, KFD(1)
DOUBLE PRECISION PREF(2,1)
INTEGER ISTOR(4), JSTOR(4), ILOAD, NN, I, J, K, L
DOUBLE PRECISION STOR(4)

101 FORMAT(110)
102 FORMAT(1H1, 'NO. OF LOAD REFERENCE CURVES = ', I5)
201 FORMAT(1H0, 'LOAD REFERENCE CURVE NO. ', 'I5/
202 FORMAT(1H0, 'NODE COMPONENT LOAD')
203 FORMAT(1H0, 'NODE Component LOAD')

NN = NF*NOD
READ(UI, 101) NLOAD
WRITE(UO, 201) NLOAD
DO 100 ILOAD = 1, NLOAD
WRITE(UO, 202) ILOAD
DO 5 I = 1, NN
5 PREF(ILOAD, I) = 0.00
11 READ(UI, 102)(ISTOR(K), JSTOR(K), STOR(K), K = 1, 4)
DO 12 K = 1, 4
12 CONTINUE
15 CONTINUE
GO TO 100
13 DO 20 K = 1, 4
20 CONTINUE
GO TO 11
100 CONTINUE
RETURN
END
SUBROUTINE READPR(UI,UC,NEL,PRREF)
C READ PRESSURE LOAD REFERENCE CURVE.
C UI,UC = INPUT,OUTPUT FILE UNIT NUMBERS.
C NEL = NUMBER OF ELEMENTS.
C PRREF(I) = INTENSITY ON ELEMENT I FOR PRESSURE REFERENCE VECTOR.
INTEGER UI,UC,NEL
DOUBLE PRECISION PRREF(1)
INTEGER ISTOR(4),ILOAD,I,K
DOUBLE PRECISION STOR(4)
101 FORMAT(I10)
102 FORMAT(4(I10,F10.0))
201 FORMAT(IH,'NO. OF PRESSURE LOAD REFERENCE CURVES =',15)
202 FORMAT(IH,'PRESSURE LOAD REFERENCE CURVE NO.',15/
1IH,'ELEMNT PRESSURE')
203 FORMAT(IH,15,3X,E12.5)
READ(UI,101)NLOAD
WRITE(UC,201)NLOAD
DO 100 ILOAD=1,NLOAD
WRITE(UC,202)ILOAD
DO 5 I=1,NEL
5 PRREF(I) = 0.00
11 READ(UI,102)(ISTOR(K),STOR(K),K=1,4)
DO 12 K=1,4
12 CONTINUE
GO TO 13
13 DO 20 K=1,4
I = ISTOR(K)
IF(I.LE.0)GO TO 20
WRITE(UC,203)I,STOR(K)
PRREF(I) = STOR(K)
20 CONTINUE
GO TO 11
100 CONTINUE
RETURN
END

SUBROUTINE READI(UI,UC,NINCR,PRED,IPRED,PFACT,PFAC1)
C READ INCREMENTAL LOAD DATA.
C UI,UO = INPUT,OUTPUT FILE UNIT NUMBERS.
C NINCR = NUMBER OF LOAD INCREMENTS.
C IPRED = DEFAULT SOLUTION PREDICTOR ORDER.
C PFAC(I,J) = NODAL LOAD FACTOR FOR INCR. I AND REFERENCE VECTOR J.
C PRFACT(I) = ELEMENT PRESSURE INTENSITY FACTOR FOR LOAD INCR. I.
INTEGER UI,UO,NINCR,IPRED(1)
DOUBLE PRECISION PFACT(2,1),PRFACT(I)

101 FORMAT(I10)
102 FORMAT(110,3F10.0)
201 FORMAT(1H1,"NO. OF LOAD INCREMENTS = ",5/ 
11H, 'INCREMENT',2X,'PREDICTOR',2X,'MECHANICAL CURVE FACTORS', 
16X,'PRESSURE')
202 FORMAT(1H1,15,6X,15,6X,E12.5,2X,E12.5,2X,E12.5)
READ(UI,101)NINCR
WRITE(UO,201)NINCR
READ(UO,202)IPRED(NINCR),PFACT(1,INCR),PFACT(2,INCR),PRFACT(INCR)
IF(IPRED(INCR).EQ.0)IPRED(INCR) = PRED
WRITE(UO,202)INCR,IPRED(INCR),PFACT(1,INCR),PFACT(2,INCR), 
PRFACT(INCR)
CONTINUE
RETURN
END

SUBROUTINE HEAD(UO,INCR,ISTEP,LAMS,LAM,FI,F2,FP, 
IPRED,MAXUP,NUP,MAXIT,ITER,ERRMAX,ERR)
C WRITE HEADING FOR LOAD INCREMENT STEP.
C UO = OUTPUT FILE UNIT NUMBER.
C INCR = LOAD INCREMENT NUMBER.
C ISTEP = LOAD STEP NUMBER.
C LAMS = LOAD STEP PARAMETER = FRACTION OF REMAINING LOAD INCREMENT.
C LAM = INCREMENTAL LOAD PARAMETER (MAXIMUM VALUE 1.0).
C FI,F2 = NODAL LOAD FACTORS APPLIED TO REFERENCE VECTORS.
C FP = ELEMENT PRESSURE LOAD FACTOR APPLIED TO REFERENCE VECTOR.
C IPRED = SOLUTION PREDICTOR ORDER.
C MAXUP = SPECIFIED MAXIMUM NUMBER OF JACOBIAN STIFFNESS UPDATES.
C NUP = NUMBER OF JACOBIAN UPDATES PERFORMED DURING THIS LOAD STEP.
MAXIT = MAXIMUM NUMBER OF RESIDUAL-FORCE ITERATIONS PER UPDATE.
ITER = NUMBER OF ITERATIONS PERFORMED SINCE LAST UPDATE.
ERRMAX = SPECIFIED MAXIMUM RESIDUAL-FORCE ERROR NORM.
ERR = ACTUAL ERROR NORM OBTAINED.

INTEGER UO, INCNCR, ISTEP, PREO, MAXUP, NUP, MAXIT, ITER
DOUBLE PRECISION LAMS, LAM, F1, F2, FP, ERRMAX, ERR

201 FORMAT(1H1, 'LOAD INCREMENT', 15,
1', LOAD STEP', 15,
2//1H, 'INCREMENT LOAD PARAMETER = ', E12.5,
3', STEP LOAD PARAMETER = ', E12.5)
202 FORMAT(1H, 'MECHANICAL LOAD FACTORS = ', E12.5)
207 FORMAT(1H, 'PRESSURE LOAD FACTOR = ', E12.5)
203 FORMAT(1H, 'PREDICTOR TYPE = ', 15)
204 FORMAT(1H, 'SPECIFIED MAX. NO. JACOBIAN UPDATES = ', 15,
1', NO. UPDATES PERFORMED = ', 15)
205 FORMAT(1H, 'SPECIFIED MAX. NO. ITERATIONS PER UPDATE = ', 15,
1', NO. ITERATIONS PERFORMED SINCE LAST UPDATE = ', 15)
206 FORMAT(1H, 'SPECIFIED MAX. RESIDUAL FORCE ERROR = ', E12.4,
1', ACTUAL ERROR = ', E12.4)

WRITE(UO, 201) INCNCR, ISTEP, LAMS
WRITE(UO, 202) F1, F2
WRITE(UO, 207) FP
WRITE(UO, 203) PREO
WRITE(UO, 204) MAXUP, NUP
WRITE(UO, 205) MAXIT, ITER
WRITE(UO, 206) ERRMAX, ERR
RETURN
END

SUBROUTINE OUTLIM(UO, NOD, NEL, NN, NF, ELNO, EGEOM, EET, CQ, P, Q, DFF, LRL, RRL, RC, RQQ, LAM, LAMR)
OUTPUT LIMIT POINT DATA.
UO = OUTPUT FILE UNIT NUMBER.
NOD, NEL = NUMBER OF NODES, ELEMENTS.
NN, NF = SYSTEM DOF, DOF PER NODE.
ELNO(J, I) = NODE NUMBERS FOR ELEMENT I.
EGEOM(J, I) = GEOMETRIC PROPERTIES FOR ELEMENT I.
EET(J, I) = PREDICTED LIMIT STRAINS FOR ELEMENT I.
CQ(I) = CUMULATIVE NODAL DISPLACEMENT FOR DOF I.
C P,Q = TEMPORARY STORAGE VECTORS.  
C DFF = FINITE-DIFFERENCE SPACING USED IN COMPUTING FORCES.  
C RL, RRL = 1ST, 2ND ORDER LOAD PARAMETER RATES.  
C RG(I), RRQ(I) = 1ST, 2ND ORDER DISPLACEMENT RATES FOR DOF I.  
C LAM = INCREMENTAL LOAD PARAMETER (MAXIMUM VALUE 1.0).  
C LAMR = LOAD YET TO BE APPLIED = 1.0 - LAM.  
INTEGER UO, NOD, NEL, N, NF, ELNO(3, 1)  
DOUBLE PRECISION EGEOM(3, 1), EET(3, 1), QQ(1), P(1), Q(1), DFF  
DOUBLE PRECISION SLIM, LAMLIM  

201 FORMAT('PREDICTED LIMIT POINT OCCURS AT LOAD INCREMENT PARAMETER = ', F12.5)  
2'THE FOLLOWING ARE PREDICTED LIMIT FORCES-DISPLACEMENTS-STRAINS')  
SLIM = -RL/RRL  
LAMLIM = LAM + (SLIM*RL + .500*SLIM**2*RRL)*LAMR  
WRITE(UO, 201) LAMLIM  
DO 10 I = 1, NN  
10 Q(I) = QQ(I) + SLIM*RQ(I) + .500*SLIM**2*RRQ(I)  
CALL FORCE(NEL, NN, NF, ELNO, DFF, Q, P)  
CALL OUTPQ(UO, NOD, NF, P, Q)  
CALL STRAIN(NEL, ELNO, EGEOM, NF, Q, EET)  
CALL OUTE(UO, NEL, EET)  
RETURN  
END  

SUBROUTINE OUTPQ(UO, NOD, NF, P, Q)  
C WRITE TOTAL FORCES AND DISPLACEMENTS.  
C UC = OUTPUT FILE UNIT NUMBER.  
C NOD, NF = NUMBER OF NODES, DOF PER NODE.  
C P(I), Q(I) = FORCE, DISPLACEMENT TO BE OUTPUT AT DOF I.  
INTEGER UO, NOD, NF, I, J  
DOUBLE PRECISION P(I), Q(I)  
DOUBLE PRECISION STOR(6)  

201 FORMAT(1H1, 14X, 22(1H*),' CUMULATIVE INTERNAL FORCES AND DISPLACEMENTS')  
1ENTS*, 2X, 23(1H*)/1H; ** NODE ***, 4X, 17(1H*), ' FORCES ', 16(1H*), 00077400  
27X, 13(1H*), ' DISPLACEMENTS ', 13(1H*), 00077500  
3/1H, 10H NO. 1.D.*, 5X, 1HU, 14X, 1HV, 14X, 1HW, 19X, 1HU, 14X, 1HV, 14X, 1HW/00077600  
1)
SUBROUTINE OUTE(UO,NEL,ET)
C WRITE CUMULATIVE STRAINS ET.
C UO = OUTPUT FILE UNIT NUMBER.
C NEL = NUMBER OF ELEMENTS.
C ET(J,I) = STRAINS TO BE OUTPUT FOR ELEMENT I.
INTEGER UO,NEL
C DOUBLE PRECISION ET(3,1)
INTEGER II, I
201 FORMAT(IH,'ELEMENT ',3X,12(IH*),' CUMULATIVE STRAINS ',12(IH*),
      1/IH, ' NO.',10X,2HXX,10X,2HYY,10X,2HZZ,10X,2HXY)
202 FORMAT(IH,'15.3X,2E12.4,12X,E12.4)
WRITE(UO,201)
DO 100 II=1,NEL
100 WRITE(UO,202)II,(ET(I,II),I=1,3)
RETURN
END

SUBROUTINE QFILL(NF,ELNO,CQ,Q)
C FORM VECTOR OF ELEMENT DISPLACEMENTS Q FROM NODAL DISPLACEMENTS QQ
C NF = DOF PER NODE.
C ELNO(J,I) = NODE NUMBERS FOR ELEMENT I.
C QQ(1) = NODAL DISPLACEMENT AT DOF I.
C Q(I) = ELEMENT NODAL DISPLACEMENT AT ELEMENT DOF I.
INTEGER NF,ELNO(I)
DOUBLE PRECISION QQ(1),Q(1)
SUBROUTINE PFILL(NF,ELNC,P,PP)
C FORM VECTOR OF NODAL FORCES PP FROM ELEMENT FORCES P.
C NF = DOF PER NODE.
C ELNC(J) = NODE NUMBERS FOR ELEMENT I.
C P(I) = ELEMENT NODAL FORCE AT ELEMENT DOF I.
C PP(I) = NODAL FORCE AT DOF I.
INTEGER NF,ELNC(I)
DOUBLE PRECISION P(1),PP(1)
CALL ROTQ(NF,ELNC,P,1)
DO 100 I=1,3
NI = ELNC(I)
IO = NF*(NI-1)
JO = NF*(I-1)
DO 100 K=1,NF
Q(JO+K) = QQ(IO+K)
100 Q(JO+K) = QQ(IO+K)
CALL ROTQ(NF,ELNC,Q,H)
RETURN
END

SUBROUTINE DFILL(NF,EGEOM,Q,D)
C COMPUTE ELEMENT DISPLACEMENT DERIVATIVES D FROM DISPLACEMENTS Q.
C NF = DOF PER NODE.
C EGEOM(J) = GEOMETRIC PROPERTIES FOR ELEMENT I.
C Q(I) = ELEMENT NODAL DISPLACEMENT AT ELEMENT DOF I.
C D(I) = ITH DISPLACEMENT DERIVATIVE (UX,VX,WX,UY,VY,WA).
INTEGER NF
DOUBLE PRECISION EGEOM#,Q(1),D(1)
INTEGER NF2, I
DOUBLE PRECISION B, H, B1, B2, A2
NF2 = NF*2.
B = EGEOM(1)
H = EGEOM(2)
B1 = EGEOM(3)
B2 = B - B1
A2 = 1.00/(B*H)
DO 10 I=1,NF
D(I) = A2*H*(-Q(I)+Q(NF+I))
10 D(NF+I) = A2*(-B2*Q(I)-B1*Q(NF+I)+B*Q(NF2+I))
RETURN
END

SUBROUTINE EFILL(NF, D, ET)
C COMPUTE ELEMENT STRAINS ET FROM DISPLACEMENT DERIVATIVES D.
C NF = DOF PER NCDE.
C C(I) = ITH DISPLACEMENT DERIVATIVE (UX, VX, WX, UY, VY, WY).
C ET(I) = ITH LAGRANGIAN STRAIN COMPONENT (XX, YY, XY).
INTEGER NF
DOUBLE PRECISION D(1), ET(1)
ET(1) = D(1)
ET(2) = D(NF+2)
ET(3) = C(2) + D(NF+1)
DO 10 I=1,NF
ET(1) = ET(1) + .500*D(I)**2
ET(2) = ET(2) + .500*D(NF+I)**2
10 ET(3) = ET(3) + D(I)*D(NF+I)
RETURN
END

SUBROUTINE AFILL(NF, D, A, KODE)
C KODE=0 FORM LAGRANGIAN A(I, J) = A1(I, J, K)*D(K).
C NF = DOF PER NODE.
C D(I) = ITH DISPLACEMENT DERIVATIVE (UX, VX, WX, UY, VY, WY).
C A GIVES LAGRANGIAN STRAINS (XX, YY, XY) FROM (UX, VX, WX, UY, VY, WY).
INTEGER NF, KODE
SUBROUTINE GFILL(NF, EGEOM, G)
C FCRM DISPLACEMENT-DERIVATIVES FROM NODAL-DISPLACEMENTS MATRIX G.
C NF = DOF PER NODE.
C EGEOM(J,1) = GEOMETRIC PROPERTIES FOR ELEMENT I.
C G GIVES (UX, VX, WX, YY, WY, WY) FROM ELEMENT NODAL DISPLACEMENTS.
INTEGER NF
DOUBLE PRECISION EGEOM(1), G(6,1)
INTEGER NF2, NF3, I, J, I1
DOUBLE PRECISION B, H, B1, B2, A2
NF2 = NF*2
NF3 = NF*3.
B = EGEOM(1)
H = EGEOM(2)
B1 = EGEOM(3)
B2 = B-B1
A2 = 1.00/(B*H)
DC 5 I=1,NF2
DC 5 J=1,NF3
G(I,J) = 0.00
DC 10 I=1,NF
I1 = NF+I
10 DC 10 J=1,NF
JL = NF+J
A(1, J) = D(J)
A(1, J1) = 0.00
A(2, J) = 0.00
A(2, J1) = D(J1)
A(3, J) = D(J1)
A(3, J1) = D(J)
IF(KODE.EQ.0) RETURN
A(1, 1) = 1.00 + A(1,1)
A(2, NF+2) = 1.00 + A(2,NF+2)
A(3, 2) = 1.00 + A(3,2)
A(3, NF+1) = 1.00 + A(3,NF+1)
RETURN
END
SUBROUTINE MTRAN(A,MA,NA,B,MB,NB,D)
C
COMPUTE C(I,J) = A(M,N)*B(M,I)*B(N,J).
A = SQUARE MATRIX TO BE TRANSFORMED.
MA = MAXIMUM (FORTRAN-DIMENSIONED) SIZE OF A.
NA = ACTUAL SIZE OF A.
B = TRANSFORMATION MATRIX.
MB = MAXIMUM (FORTRAN-DIMENSIONED) SIZE OF B.
NB = ACTUAL SIZE OF B.
INTEGER MA,NA,MB,NB
COUPLE PRECISION A(MA,1),B(MA,1),D(MB,1)
INTEGER I,J,M
DOUBLE PRECISION C,STOR(6,9)
D0 50 I=1,NA
DC 50 J=1,NB
C = 0.DO
DO 45 M=1,NA
45 C = C + A(I,M)*B(M,J)
50 STOR(I,J) = C
DO 100 I=1,NB
DC 100 J=1,NB
C = 0.DO
DO 95 M=1,NA
95 C = C + STOR(M,J)*B(M,I)
100 C(I,J) = C
RETURN
END

SUBROUTINE ROTQ(NF,ELNC,Q,KODE)
C KODE=0 ROTATE DISPLACEMENTS TO ELEMENT FROM NODAL.
C KODE=1 ROTATE FORCES TO NODAL FROM ELEMENT.
C NF = DOF PER NODE.
C ELNO(J,1) = NODE NUMBERS FOR ELEMENT 1.
C Q(I) = ELEMENT NODAL DISPLACEMENT AT ELEMENT DOF I.
C GCOS(J,1) = DIRECTION COSINES FOR COORDINATE SYSTEM AT NODE I.
C CCORD(J,1) = COORDINATES OF NODE I.
C COMMON/COMCOS/GCOS/COMCRR/CCORD
INTEGER NF,ELNO(l),KODE
DOUBLE PRECISION GCOSI3,3,I),COORD(3,1),Q(1)
INTEGER N1,N2,N3,N1,N1,J,M,I0
DOUBLE PRECISION V21(3),V31(3),VY(3),VZ(3)
DOUBLE PRECISION C,REG(3,3),RNG(3,3),REM(3,3),QPART(3)
C COMPUTE MATRIX REG TO ROTATE DISPLACEMENTS ELEMENT FROM GLOBAL
N1 = ELNO(1)
N2 = ELNO(2)
N3 = ELNO(3)
DO 10 I=1,3
V21(I) = CCORD(I,N2) - CCORD(I,N1)
10 V31(I) = COORD(I,N3) - CCORD(I,N1)
CALL VCROSS(V21,V31,VZ)
CALL VNCRM(V21,V21)
CALL VNORM(VZ,VZ)
CALL VCROSS(VZ,V21,VY)
DO 20 J=1,NF
REG(I,J) = V21(J)
20 REG(3,J) = VZ(J)
DO 50 N=1,3
C COMPUTE MATRIX RNG ROTATE NODE N DISPLACEMENTS NODAL FROM GLOBAL
NI = ELNO(N)
DO 30 I=1,NF
DO 30 J=1,NF
30 RNG(I,J) = GCOS(I,J,NI)
C COMPUTE MATRIX REN ROTATE DISPLACEMENTS ELEMENT FROM NODAL
DO 50 I=1,NF
DO 50 J=1,NF
C = 0.DO
DO 45 M=1,NF
45 C = C + REG(I,M)*RNG(J,M)
SUBROUTINE ROTK(NF,ELN0,K)
C RCTATE ELEMENT STIFFNESS TO NODAL FROM ELEMENT.
C NF = DOF PER NODE.
C ELNO(J,I) = NODE NUMBERS FOR ELEMENT I.
C K = ELEMENT STIFFNESS MATRIX.
C GCOS(J,I) = DIRECTION COSINES FOR COORDINATE SYSTEM AT NODE I.
C CCORD(J,I) = COORDINATES OF NODE I.
COMMON/GCOS/GCOS/COORD/COORD
INTEGER NF,ELN0(1)
DOUBLE PRECISION GCOS(3,3,1),CCORD(3,1),K(9,9)
INTEGER N1,N2,N3,N,NI,N,J,K,10,J0,IP,JP
DOUBLE PRECISION V21(3),V31(3),VY(3),VZ(3)
DOUBLE PRECISION C,REG(3,3),RNG(3,3),REN(3,3),KPART(3,3)
C COMPUTE MATRIX REG TO ROTATE DISPLACEMENTS ELEMENT FROM GLOBAL
N1 = ELNO(1)
N2 = ELNO(2)
N3 = ELNO(3)
DO 10 I=1,3
V21(I) = CCORD(I,N2) - CCORD(I,N1)
10 V31(I) = CCORD(I,N3) - CCORD(I,N1)
CALL VCROSS(V21,V31,VZ)
CALL VNORM(V21,V21)
CALL VNORM(VZ,VZ)
CALL VCROSS(VZ,V21,VY)
DO 20 J=1,NF
  REG(1,J) = V21(J)
  REG(2,J) = VY(J)
20  REG(3,J) = VZ(J)
DO 500 N=1,3
C COMPUTE MATRIX RNG ROTATE NODE N DISPLACEMENTS NODAL FROM GLOBAL
NI = ELNO(N)
DO 30 I=1,NF
  CC 30 J=1,NF
  30 RNG(I,J) = GCOS(I,J,NI)
C COMPUTE MATRIX REN ROTATE DISPLACEMENTS ELEMENT FROM NODAL
DO 50 I=1,NF
  CC 50 J=1,NF
  50 REN(I,J) = C
DO 200 IP=1,3
C Rotate K(IP,N) PARTITION USING MATRIX REN FOR NODE N
10 = NF*(IP-1)
JO = NF*(N-1)
DO 110 I=1,NF
  DO 110 J=1,NF
  KPART(I,J) = K(IO+I,JO+J)
110 CONTINUE
DO 300 JP=1,3
C Rotate K(N,JP) PARTITION USING MATRIX REN FOR NODE N
10 = NF*(N-1)
JO = NF*(JP-1)
DO 210 I=1,NF
  DO 210 J=1,NF
  K(IO+I,JO+J) = C
210 CONTINUE
SLBROUTINE FORCE(NEL,NN,NF,ELNO,DF,QQ,PP)
C COMPUTE INTERNAL FORCES PP CORRESPONDING TO DISPLACEMENTS QQ.
C NEL = NUMBER OF ELEMENTS.
C NN, NF = SYSTEM DOF, DOF PER NODE.
C ELNO(I) = NODE NUMBERS FOR ELEMENT I.
C DF = FINITE-Difference SPACING USED IN COMPUTING FORCES.
C QQ(I), PP(I) = CUMULATIVE DISPLACEMENT, INTERNAL FORCE AT DOF I.
C EGEOM(I) = GEOMETRIC PROPERTIES FOR ELEMENT I.
C T(I) = THICKNESS OF ELEMENT I.
C NS = NUMBER OF STRAIN COMPONENTS.
COMMON/COMEG/EGEOM/COMT/COMNS/NS
INTEGER NEL, NN, NF, ELNO(3,1), NS
INTEGER III, M, NF2, NF3
DOUBLE PRECISION CV, UA, UB, EO(3), E(3), S(3), D(6), PD(6), Q(9), P(9)
DOUBLE PRECISION A(3,6), G(6,9), ENERGY
EQUIVALENCE(A(I), G(I))
NF2 = NF*2
NF3 = NF*3
DO 10 I=I,NN
PP(I) = 0.DO
DO 100 II=1,NEL
V = .5DO*TI**II*I*EGEOM(1,II)*EGEOM(2,II)
CALL WFFILL(NF,ELNO(I,II),Q,C)
CALL DFFILL(NF,EGEOM(I,II),Q,D)
CALL EFILL(NF,D,EO)
DO 50 I=1,NS
DO 35 M=1,NS
 35 E(M) = E0(M)
   C = E0(I)
   E(I) = C - DF
   UA = ENERGY(I,E)
   E(I) = C + DF
   UB = ENERGY(I,E)
50 S(I) = (UB-UA)/(2.0*DF)*V
   CALL AFILL(NF,D,A,1)
   DO 70 I=1,NF2
     C = 0.DO
   DO 65 M=1,NS
65 C = C + A(M,I)*S(M)
70 PD(I) = C
   CALL GFILL(NF,EIGEOM(I,II),G)
   DO 80 I=1,NF3
     C = 0.DO
   DO 75 M=1,NF2
75 C = C + PD(M)*G(M,I)
80 P(I) = C
100 CALL PFILL(NF,ELNO(I,II),P,PP)
RETURN
END

SUBROUTINE PFORCE(PFACT,PREF,NN,P)
  C COMPUTE APPLIED NODAL FORCES P.
  C PFACT(J) = NODAL LOAD FACTOR FOR REFERENCE VECTOR J.
  C PREF(J,I) = NODAL LOAD AT DOF I FOR REFERENCE VECTOR J.
  C NN = TOTAL SYSTEM DOF.
  C P(I) = APPLIED CUMULATIVE LOAD AT DOF I.
  INTEGER NN
  DOUBLE PRECISION PFACT(1),PREF(2,1),P(1)
  INTEGER I
  DO 100 I=1,NN
100 P(I) = PFACT(1)*PREF(1,I) + PFACT(2)*PREF(2,I)
RETURN
END
SUBROUTINE EFORCE(IPRESS,PR,PRREF,CQ,NEL,NN,NF,ELNC,PP)
C COMPUTE NODAL FORCES PP DUE TO ELEMENT PressURES PR.
C IPRESS = NONCONSERVATIVE CODE = 0,1 FOR NO PRESSURE, PRESSURE.
C PR = ELEMENT PRESSURE INTENSITY FACTOR.
C PRREF(I) = INTENSITY ON ELEMENT I FOR PRESSURE REFERENCE VECTOR.
C CQ(I) = CUMULATIVE NODAL DISPLACEMENT AT DOF I.
C NEL = NUMBER OF ELEMENTS.
C NN,NF = TOTAL SYSTEM DOF, DOF PER NODE.
C ELNO(J,I) = NODE NUMBERS FOR ELEMENT I.
C PP(I) = COMPUTED NONCONSERVATIVE PRESSURE NODAL FORCE AT DOF I.
C EGEOM(J,I) = GEOMETRY FOR ELEMENT I (BASE, HEIGHT, PART BASE).
COMMON/COMEG/EGEOM
INTEGER IPRESS,NEL,NN,NF,ELNO(3,1)
DOUBLE PRECISION PR,PRREF(1),CQ(1),PP(1),EGEOM(3,1)
INTEGER I,II
DOUBLE PRECISION C,V21(3),V31(3),Q(9),P(9)
DO 10 I=1,NN
10 IF(IPRESS.EQ.0)RETURN
DO 100 II=1,NEL
CALL QFILL(NF,ELNO(1,II),Q
call 20 I=1,3
20 V21(I) = Q(3+I) - Q(I)
20 V31(I) = Q(6+I) - Q(I)
V21(1) = V21(1) + EGEOM(1,II)
V31(1) = V31(1) + EGEOM(3,II)
V31(2) = V31(2) + EGEOM(2,II)
CALL VCROSS(V21,V31,P)
C = PR*PRREF(II)/6.DO
DO 50 I=1,3
50 P(I) = P(I)*C
P(3+I) = P(I)
P(6+I) = P(I)
100 CALL PFILL(NF,ELNO(1,II),P,PP)
RETURN
END

SUBROUTINE ERCOMP(UO,NN,KFD,PP,P,ERR)
C COMPUTE ERROR NORM USING CUMULATIVE FORCES PP AND RESIDUALS P.
UC = OUTPUT FILE UNIT NUMBER.

NN = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.

KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.

PP(I) = CUMULATIVE NODAL FORCE AT DOF I.

P(I) = RESIDUAL (UNBALANCED) NODAL FORCE AT DOF I.

ERR = COMPUTED RESIDUAL-FORCE ERROR NORM.

INTEGER UD,NN,KFD(I)

DOUBLE PRECISION PP(1),P(1),ERR

INTEGER I,J

DOUBLE PRECISION C1,C2

201 FORMAT(IH,'ERROR NORM=',E12.5)

ERR = 0.00

C1 = 0.00

C2 = 0.00

DC 5 I=1,NN

J = KFD(I)

IF(J.LT.0.OR.J.EQ.I)GO TO 5

P(J) = P(J) + P(I)

P(I) = 0.00

5 CONTINUE

DC 10 I=1,NN

IF(KFD(I).EQ.I)C1 = C1 + DABS(P(I))

10 C2 = C2 + DABS(PP(I))

IF(C2.GT.0.DO)ERR = C1/C2

WRITE(UD,201)ERR

RETURN

END

SUBROUTINE STRAIN(NEL,ELNO,EGEOM,NF,QQ,EET)

COMPUTE STRAINS EET FROM GLOBAL DISPLACEMENTS QQ.

NEL = NUMBER OF ELEMENTS.

ELNO(J,I) = NODE NUMBERS FOR ELEMENT I.

EGEOM(J,I) = GEOMETRIC PROPERTIES FOR ELEMENT I.

NF = DOF PER NODE.

CC(I) = CUMULATIVE NODAL DISPLACEMENT AT DOF I.

EET(J,I) = COMPUTED CUMULATIVE STRAINS (XX,YY,XY) FOR ELEMENT I.

INTEGER NEL,ELNO(3,1),NF

DOUBLE PRECISION EGEOM(3,1),QQ(1),EET(3,1)

INTEGER I
DOUBLE PRECISION Q(9), D(6)
DO 10 II=1,NEL
CALL WFill(NF,ELNO(1,II),QQ,Q)
CALL DFILL(NF,Egeom(1,II),Q,D)
10 CALL EfILL(NF,D,EET(1,II))
RETURN
END

DOUBLE PRECISION FUNCTION ENERGY(II,ET)
C EVALUATE ENERGY DENSITY FOR ELEMENT II AT STRAINS ET (MCONEY).
C II = ELEMENT NUMBER.
C ET(I) = CUMULATIVE STRAINS (XX,YY,XY) FOR ELEMENT.
C IMAT(I) = MATERIAL NUMBER FOR ELEMENT I.
C CC1(I),CC2(I) = MATERIAL PROPERTIES FOR MATERIAL I.
COMMON/COMMAT/IMAT/COMENU/CC1,CC2
INTEGER II,IMAT(I)
DOUBLE PRECISION ET(.II),CC(5),CC(5)
INTEGER I
DOUBLE PRECISION C1,C2,A,B,D,I1,I2
I = IMAT(II)
C1 = CC1(I)
C2 = CC2(I)
A = 2.DO*ET(1)+ET(2))
B = 4.DO*ET(1)*ET(2)-ET(3)**2
D = 1.DO + A + B
I1 = (A*(A+B)-B)/D
I2 = I1 + B*(A+B)/D
ENERGY = C1*I1 + C2*I2
RETURN
END

SUBROUTINE EVAL(II,ORD,N,FO,DF,ESTOR)
C EVALUATE STRAIN ENERGY OF ELEMENT II AS FUNCTION OF STRAINS TO
C ESTABLISH A COMPLETE INTERPOLATING POLYNOMIAL OF ORDER ORD L.E. 3.
C II = ELEMENT NUMBER.
C ORD = MAXIMUM TENSOR ORDER TO BE USED FOR DIFFERENCE EXPANSION.
C N = DIMENSION OF TENSORS.
C FO(I) = CURRENT VALUE OF INDEPENDENT VARIABLE I.

C CF(I) = FINITE DIFFERENCE IN INDEPENDENT VARIABLE I.
C ESTOR(I) = STORAGE VECTOR FOR ENERGY EVALUATIONS I.
C ENERGY(I,I,F) GIVES ENERGY FOR ELEMENT II AT VARIABLE STATE F.
INTEGER II, ORD, N
DOUBLE PRECISION FO(1), DF(1), ESTCR(I), ENERGY
INTEGER I, J, K, M, IE
DOUBLE PRECISION F(3)
IE = 1
ESTOR(IE) = ENERGY(II,FO)
IF(ORD .LT. 1) GO TO 100
DO 10 I=1,N
IE = IE+1
DO 5 M=1,N
5 F(M) = FO(M)
F(I) = F(I) + DF(I)
ESTOR(IE) = ENERGY(II,F)
IF(ORD .LT. 2) GO TO 100
DO 10 J=1,I
DO 15 K=1,J
15 F(M) = FO(M)
F(I) = F(I) + DF(I)
F(J) = F(J) + DF(J)
F(K) = F(K) + DF(K)
ESTOR(IE) = ENERGY(II,F)
IF(ORD .LT. 3) GO TO 100
DO 30 I=1,N
DO 30 J=1,I
DO 30 K=1,J
30 F(M) = FO(M)
F(I) = F(I) + DF(I)
F(J) = F(J) + DF(J)
F(K) = F(K) + DF(K)
ESTOR(IE) = ENERGY(II,F)
100 RETURN
END
SUBROUTINE U2FORM(N, DF, ESTOR, USTOR)

C FORM STRAIN ENERGY TENSORS USTOR USING QUADRATIC POINTS ESTOR.

C N = DIMENSION OF TENSORS.
C
C DF(I) = FINITE DIFFERENCE IN INDEPENDENT VARIABLE I.
C ESTOR(I) = STORAGE VECTOR FOR ENERGY EVALUATION I.
C USTOR(I) = STORAGE VECTOR FOR TENSOR COMPONENT I.

INTEGER N

DOUBLE PRECISION DF(1), ESTOR(1), USTOR(1)

INTEGER I(1, 1, 1, 1, 1, 1, 1), LOCII, NI, NJ

DOUBLE PRECISION C, CI, CJ, CIJ, DI, DJ

NI = 1 + N

II = 1

I2 = N1

C FORM 0TH-ORDER TENSOR

C = ESTOR(1)

USTOR(1) = C

C FORM 1ST-ORDER TENSOR

DO 100 I = 1, N

DI = DF(I)

I1 = I1 + 1

CI = ESTOR(I1)

LOCII = NI + I*(I + 1)/Z

C FORM X TYPE TERM

USTOR(II) = (-1.500*C + 2.00*CI - 0.500*ESTOR(LOCII))/DI

C FORM 2ND-ORDER TENSOR

DO 100 J = 1, I

CJ = DI*DF(J)

I2 = I2 + 1

CIJ = ESTOR(I2)

IF(J.LT.I) GO TO 79

C FORM XX TYPE TERM

USTOR(I2) = (C - 2.00*CI + CIJ)/DJ

GO TO 100

79 CJ = ESTOR(I + J)

C FORM YX TYPE TERM

USTOR(I2) = (C - CI - CJ + CIJ)/DJ

100 CONTINUE

RETURN

END
SUBROUTINE U3FORM(N,DF,ESTOR,USTOR)  
C FORM STRAIN ENERGY TENSORS USTOR USING CUBIC POINTS ESTOR.  
C N = DIMENSION OF TENSORS.  
C DF(I) = FINITE DIFFERENCE IN INDEPENDENT VARIABLE I.  
C ESTOR(I) = STORAGE VECTOR FOR ENERGY EVALUATIONS I.  
C USTOR(I) = STORAGE VECTOR FOR TENSOR COMPONENT I.  
INTEGER N  
DOUBLE PRECISION DF(I),ESTOR(I),USTOR(I)  
INTEGER I,J,K,I1,I2,I3,LOCII,LOCJJ,LOCIII,LOCIIJ,LOCIIJJ,LOCJK,  
1LOCJK,N1,N2  
DOUBLE PRECISION C,C116,C13,C11,C13*ESTOR(LOCIII) + 3.00*CI  
1- 1.500OESTOR(LOCII))/DI  
C FORM X TYPE TERM  
C FORM 2ND-ORDER TENSOR  
INTEGER I,J,K,I1,I2,I3,LOCII,LOCJ,LOCIII,LOCIIJ,LOCJJ,LOCJK,  
1LOCJK,N1,N2  
DOUBLE PRECISION C,C116,C13,C11,C13*ESTOR(LOCIII) + 3.00*CI  
1- 1.5000ESTOR(LOCII))/DI  
C FORM XX TYPE TERM
SUBROUTINE UFILL(II,ORD,ETUSTOR)

C FILL USTOR OF ORDER ORD FOR ELEMENT II.

C II = ELEMENT NUMBER.

C ORD = MAXIMUM TENSOR ORDER TO BE USED FOR DIFFERENCE EXPANSION.

C ET(I) = CURRENT VALUE OF STRAIN COMPONENT I.

C USTOR(I) = STORAGE VECTOR FOR TENSOR COMPONENT I.

C NS = NUMBER OF STRAIN COMPONENTS.

C CFE = FINITE DIFFERENCE SIZE FOR STRAIN VARIABLES.

USTOR(I2) = (2.00*C - ESTOR(LOCIII) - 5.00*CI + 4.00*CIJ)/DJ

GO TO 81

LOCJ = N1 + J*(J+1)/2
LOCIIJ = N2 + (I-1)*I*(I+1)/6 + (J+1)*J/2
LOCIIIJ = N3 + (I-1)*I*(I+4)/6 + J

C FORM YX TYPE TERM
USTOR(I2) = (2.00*C - 2.500*CI + 3.00*CIJ)/DJ

GO TO 81

LOCIIJ = N2 + (I-1)*I*(I+4)/6 + J

C FORM 3RD-ORDER TENSOR

1

GO 100 K=1, J

CK = DJ*DF(K)

I3 = I3+1

CIJK = ESTOR(I3)

LOCJK = N1 + (J-1)*J/2 + K

C FORM XXX TYPE TERM

IF(K.EQ.J) USTOR(I3) = (-C + CIJK + 3.00*CI - 3.00*CIJ)/DK

GO TO 100

CONTINUE

RETURN

END
COMMON/COMNS/NS/COMDFE/DFE
INTEGER II, ORD, NS
DOUBLE PRECISION ET(1), USTOR(1), DFE
INTEGER I
DOUBLE PRECISION ESTOR(20), DF(3)
DO 10 I = 1, NS
   DF(I) = DFE
   CALL EVAL(II, ORD, NS, ET, DF, ESTOR)
   IF (ORD .EQ. 2) CALL U2FORM(NS, DF, ESTOR, USTOR)
   IF (ORD .EQ. 3) CALL U3FORM(NS, DF, ESTOR, USTOR)
RETURN
END

SUBROUTINE CFORMIQ, EGEOM, CMAT
C FORM 3 X 9 PRESSURE LOAD MATRIX CMAT.
C Q(I) = CUMULATIVE NODAL DISPLACEMENT AT DOF I.
C EGEOM(J, I) = GEOMETRIC PROPERTIES FOR ELEMENT I.
C CMAT = UNSYMMETRIC STIFFNESS PARTITION DUE TO PRESSURE LOADS.
DOUBLE PRECISION Q(1), EGEOM(1), CMAT(3, 9)
INTEGER I
DOUBLE PRECISION CX2, CY2, CZ2, CX3, CY3, CZ3
CX2 = EGEOM(1) + Q(4) - Q(1)
CY2 = Q(5) - Q(2)
CZ2 = Q(6) - Q(3)
CX3 = EGEOM(3) + Q(7) - Q(1)
CY3 = EGEOM(2) + Q(8) - Q(2)
CZ3 = Q(9) - Q(3)
DO 10 I = 1, 3
   CMAT(I, I) = 0.00
   CMAT(I, I + 3) = 0.00
   CMAT(I, I + 6) = 0.00
   CMAT(I, 2) = CZ2 - CZ3
   CMAT(I, 3) = -CY2 + CY3
   CMAT(I, 5) = CZ3
   CMAT(I, 6) = -CY3
   CMAT(I, 8) = -CZ2
   CMAT(I, 9) = CY2
   CMAT(2, 1) = -CZ2 + CZ3
   CMAT(2, 3) = CX2 - CX3
SUBROUTINE GENER8(II,K)
C  FORM STIFFNESS MATRIX K FOR ELEMENT II IN NODAL COORDINATES.
C II,K = ELEMENT NUMBER,ELEMENT STIFFNESS MATRIX.
C NS,NF = NUMBER OF STRAIN COMPONENTS,DOF PER NODE.
C ELNO(J,I) = NODE NUMBERS FOR ELEMENT I.
C EGEOM(J,I) = GEOMETRIC PROPERTIES FOR ELEMENT I.
C T(I) = THICKNESS OF ELEMENT I.
C CQII) = CUMULATIVE NODAL DISPLACEMENT AT DOF I.
C IPRESS = NONCONSERVATIVE CODE = 0,1 FOR NO PRESSURE,PRESSURE.
C PR = ELEMENT PRESSURE INTENSITY FACTOR.
C PRREF(I) = INTENSITY ON ELEMENT I FOR PRESSURE REFERENCE VECTOR.
COMMON/COMNS/NS/NF/COMEL/ELNO/COMEG/EGEOM/COMT/T/COMQQ/QQ
COMMON/COMIPR/IPRESS/CCMPR/PR/CCMPRR/PRREF
INTEGER II,NS,NF,ELNO(31),IPRESS
DOUBLE PRECISION K(9,9),EGEOM(31),T(1),CQ(1),PR,PRREF(1)
DOUBLE PRECISION V,E(3),S(3),D(6),Q(9),A(3,6),
1H(6,6),G(6,6),USTOR(10)
DOUBLE PRECISION C,P6,CMAT(3,9)
EQUIVALENCE(A(1),G(1))
V = .5DOT(T(11)*EGEOM(1,II)*EGEOM(2,II))
NF2 = NF*2
NF3 = NF*3
CALL QFILL(NF,ELNO(1,II),QQ,Q)
CALL DFILL(NF,EGEOM(1,II),Q,D)
CALL EFILL(NF,0,E)
CALL UFILL(I1,2,E,USTOR)
IU = 1
DO 10 I=1,NS
IU = IU+1
10 S(I) = USTOR(IU)*V
IU = IU+1
DO 20 I=1,NS
IU = IU+1
20 DO(J,I) = DO(I,J)
CALL AFILL(NF,D,A,I)
CALL MTRAN(DO,3,NS,A,6,NF2,H)
DO 50 I=1,NF
IU = NF+I
11 H(I,1) = H(I,1) + S(1)
12 H(I,11) = H(I,11) + S(2)
13 H(I,11) = H(I,11) + S(3)
50 H(I,11) = H(I,11) + S(3)
CALL GFILL(NF,E,EGEOM(I1,II),G)
CALL MTRAN(H,6,NF2,G,9,NF3,K)
IF(IPRESS.EQ.0)GO TO 101
CALL CFORM(Q,EGEOM(I1,II),CMAT)
P6 = PR*PRREF(I1)/6.DO
DO 100 I=1,NF
100 C = P6*CMAT(I,J)
K(I,J) = K(I,J) - C
K(I+3,J) = K(I+3,J) - C
K(I+6,J) = K(I+6,J) - C
101 CALL ROTK(NF,ELNO41,I),K
RETURN
END

SUBROUTINE USUML(ORD,USTOR,N,Q,P)
C FOR ORD=2 COMPUTE P(I) = USTOR(I,J)*Q(J).
C FOR ORD=3 COMPUTE P(I) = USTOR(I,J,K)*Q(J)*Q(K).
C ORD = TENSCR ORDER TO BE USED FOR FINITE-DIFFERENCE EXPANSIONS.
C USTOR = TENSOR STORAGE ARRAY.
C N = TENSOR DIMENSION.
Q = TENSOR VECTOR ARGUMENT.
P = COMPUTED SUMMED VECTOR.
INTEGER ORD,N
DOUBLE PRECISION USTOR(I),P(I),Q(I)
INTEGER I,J,K,IU
DOUBLE PRECISION Q1,QJ,QK,QIJ,C
CC 100 I=1,N
100 P(I) = 0.00
   IU = 0
   IF(ORD.GT.2)GO TO 201
   DO 200 I=1,N
   Q1 = Q(I)
   DO 200 J=1,I
   IU = IU+1
   C = USTOR(IU)
   P(I) = P(I) + C*Q(J)
   IF(J.NE.I)P(J) = P(J) + 2.00*C*QIJ
200 CONTINUE
   RETURN
201 DO 300 I=1,N
   Q1 = Q(I)
   DO 300 J=1,I
   QJ = Q(J)
   QIJ = Q1*QJ
   DO 300 K=1,J
   QK = Q(K)
   IU = IU+1
   C = USTOR(IU)
   P(I) = P(I) + C*QJ*QK
   IF(K.EQ.J)GO TO 61
   P(I) = P(I) + C*QIJ
   GO TO 71
61 IF(J.EQ.I)GO TO 300
   P(J) = P(J) + 2.00*C*QIJ
   GO TO 300
71 P(K) = P(K) + C*QIJ
   IF(J.EQ.I)GO TO 300
   P(J) = P(J) + 2.00*C*QIJ
   P(K) = P(K) + C*QIJ
300 CONTINUE
SUBROUTINE USUM21(USTOR,N,Q,Q2,P)  
COMPUTE P(I) = USTOR(I,J,K)*Q1(J)*Q2(K).  
USTOR = TENSOR STORAGE ARRAY. 
N = TENSOR DIMENSION. 
Q1,Q2 = TENSOR VECTOR ARGUMENTS. 
P = COMPUTED SUMMED VECTOR. 
INTEGER N  
DOUBLE PRECISION USTOR(I),Q1(1),Q2(1),P(I)  
INTEGER I,J,K,IU 
DOUBLE PRECISION C 
DC 100 I=1,N  
100 P(I) = 0,DO 
   IU = 0 
   DO 200 I=1,N 
   DO 200 J=1,I 
   DO 200 K=1,J 
   IU = IU+1 
   C = USTOR(IU) 
   P(I) = P(I) + C*Q1(J)*Q2(K) 
   IF(K.EQ.J)GO TO 161 
   P(I) = P(I) + C*Q1(K)*Q2(J) 
   GO TO 171 
161 IF(J.EQ.I)GO TO 200 
   P(J) = P(J) + C*(Q1(I)*Q2(J)+Q1(J)*Q2(I)) 
   GO TO 200 
171 P(K) = P(K) + C*Q1(I)*Q2(J) 
   IF(J.EQ.I)GO TO 200 
   P(J) = P(J) + C*(Q1(I)*Q2(K)+Q1(K)*Q2(I)) 
   P(K) = P(K) + C*Q1(J)*Q2(I) 
200 CONTINUE 
RETURN 
END 

SUBROUTINE PICOMP(NN,QQSTAR,QQDOT,PR0,PR1,PP)  
COMPUTE 2ND ORDER FUNDAMENTAL LOAD TERM PP USING FUNDAMENTAL
DISPLACEMENTS (REFERENCE VALUES QQSTAR AND RATES QQDOT).

NN = TOTAL SYSTEM DOF.

QQSTAR(I) = CURRENT CUMULATIVE NODE DISPLACEMENT AT NODE I.

QQDOT(I) = NODEAL DISPLACEMENT RATE AT NODE I.

PRO,PR1 = PRESSURE FACTORS AT START,END OF LOAD STEP.

PP(I) = COMPUTED DUED FORCE TERM AT DOF I.

ORD = MAXIMUM TENSOR ORDER TO BE USED FOR DIFFERENCE EXPANSION.

NEL,NS,NF = NUMBER OF ELEMENTS, STRAIN COMPONENTS, DOF PER NODE.

ELNO(I,J) = NODE NUMBERS FOR ELEMENT I.

EGEO(I,J) = GEOMETRIC PROPERTIES FOR ELEMENT I.

T(I) = THICKNESS OF ELEMENT I.

IPRESS = NONCONSERVATIVE CODE = 0,1 FOR NO PRESSURE, PRESSURE.

PRREF(I) = PRESSURE INTENSITY ON ELEMENT I FOR PRESSURE VECTOR.

COMMON/COMRD/ORD/COMNEL/NEL/COMNS/NS/COMNF/NF
COMMON/COMEL/ELNO/COMEG/EGEO/CONT/T
COMMON/COMIPR/IPRESS/COMPRR/PRREF
INTEGER NN,ORD,NEL,NS,NF,ELNO(3,1),IPRESS

COMPUTE QQSTAR(1),QQDOT(1),PRO,PR1,PP(1),EGEO(3,1),T(1),IPRESS(1)
INTEGER NF2,NF3,IU2,IU3,II,I,M

DOUBLE PRECISION QQSTAR(9),QQDOT(9),DSTAR(6),DDOT(6),ESTAR(3),
1EDOT(3),ASTAR(3,6),AI(DOT(3),6),STCRA(3),STORB(3),STCRM(6,9),
2P(9),USTAR(20)

DOUBLE PRECISION CO,C1,VSTAR(6),V1DOT(6),V1(3),V2(3),V3(3)

NF2 = NF*2
NF3 = NF*3
IU2 = 2 + NS
IU3 = IU2 + NS*(NS+1)/2

DO 5 I=1,NN
5 PP(I) = 0.00

DO 100 II=1,NEL
V = .5*D0*T(II)*EGEO(M,1,II)*EGEO(M,2,II)

FORM FUNDAMENTAL REFERENCE QUANTITIES
CALL QFILL(NF,ELNO(1,II),QQSTAR,GSTAR)
CALL DFILL(NF,EGEO(M,1,II),QQSTAR,DSTAR)
CALL AFILL(NF,DSTAR,ASTAR,1)
CALL EFILL(NF,DSTAR,ESTAR)

FORM STRESS-STRAIN TENSORS
CALL UFILL(NF,T,ORD,ESTAR,USTAR)

FORM FUNDAMENTAL RATE QUANTITIES
CALL QFILL(NF,ELNO(1,II),QQDCT,QDCT)  
CALL DFILL(NF,EGEOM(1,II),QDOT,DDOT)  
CALL AFILL(NF,CDOT,AIDOT,0)  
CC 30 I=1,NS  
C = 0.DO  
DO 25 M=1,NF2  
25 C = C + ASTAR(I,M)*DDOT(M)  
ECOT(I) = C  
FORM STORM = DO PORTION OF DISPLACEMENT DERIVATIVE PSUEDO FORCES  
CALL USUM1(2,USTAR(IU2),NS,EDOT,STORA)  
CC 110 I=1,NF2  
C = 0.DO  
DO 105 M=1,NS  
105 C = C + STORA(M)*A1DOT(M,I)  
110 STORM(I) = 2.DO*C  
DO 114 I=1,NS  
C = 0.DO  
DO 112 M=1,NF2  
112 C = C + A1DOT(I,M)*DDOT(M)  
114 STORB(I) = C  
CALL USUM1(2,USTAR(IU2),NS,STORB,STORA)  
DO 120 I=1,NF2  
C = 0.DO  
DO 115 M=1,NS  
115 C = C + STORA(M)*ASTAR(M,I)  
120 STORM(I) = STORM(I) + C  
IF(ORD.LT.3)GO TO 501  
ADD DI PORTION OF DISPLACEMENT DERIVATIVE PSUEDO FORCES TO STORM  
CALL USUM1(3,USTAR(IU3),NS,EDOT,STORA)  
DO 210 I=1,NF2  
C = 0.DO  
DO 205 M=1,NS  
205 C = C + STORA(M)*ASTAR(M,I)  
210 STORM(I) = STORM(I) + C  
C COMPUTE ELEMENT FORCES P(I) = VOLUME INTEGRAL OF G(M,I)*STORM(M)  
501 CALL GFILL(NF,EGEOM(1,II),G)  
DO 510 I=1,NF3  
C = 0.DO  
DO 505 M=1,NF2  
505 C = C + G(M,I)*STORM(M)
510 P(I) = C*V
IF(IPRESS.EQ.0)GO TO 1000
C ADD NONCONSERVATIVE PRESSURE FORCES TO P
VSTAR(1) = EGEOM(1,II) + QSTAR(4) - QSTAR(1)
VSTAR(2) = QSTAR(5) - QSTAR(2)
VSTAR(3) = QSTAR(6) - QSTAR(3)
VSTAR(4) = EGEOM(3,II) + QSTAR(7) - QSTAR(1)
VSTAR(5) = EGEOM(2,II) + QSTAR(8) - QSTAR(2)
VSTAR(6) = QSTAR(9) - QSTAR(3)
VDOT(1) = QDOT(4) - QDOT(1)
VDOT(2) = QDOT(5) - QDOT(2)
VDOT(3) = QDOT(6) - QDOT(3)
VDOT(4) = QDOT(7) - QDOT(1)
VDOT(5) = QDOT(8) - QDOT(2)
VDOT(6) = QDOT(9) - QDOT(3)
C0 = PRO*PRREF(II)/3.DO
C1 = (PRI-PRO)*PRREF(II)/3.DO
CALL VCROSS(VDOT,VSTAR(4),V1)
CALL VCROSS(VSTAR,VDOT(4),V2)
CALL VCROSS(VDOT,VDOT(4),V3)
C 610 I=1,3
C = C1*(V1(I)+V2(I)) + C0*V3(I)
P(I) = P(I) - C
P(I+3) = P(I+3) - C
P(I+6) = P(I+6) - C
C ADD ELEMENT FORCES P TO SYSTEM FORCES PP
1000 CALL PFILL(NF,ELNC(1,II),P,PP)
RETURN
END

SUBROUTINE RATES(KMAT,P1,NN,KFD,RP,PRO,PRI,LSIGN,RL,RRL,RQ,RRQ)
C COMPUTE 1ST AND 2ND PATH RATES FOR LOAD DIRECTION VECTOR RP.
C LOAD PARAMETER RATES = RL,RRL. DISPLACEMENT RATES = RQ,RRQ.
C KMAT = SYSTEM JACOBIAN STIFFNESS MATRIX.
C P1 = TEMPORARY FORCE STORAGE VECTOR.
C NN = TOTAL SYSTEM DOF.
C KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.
C RP(I) = RESIDUAL LOAD (LOAD STEP) AT DOF I.
C PRO,PRI = PRESSURE FACTORS AT START,END OF LOAD STEP.
C LSIGN = +,- IF LOAD PARAMETER IS INCREASING, DECREASING.
C
C RL, RRL = COMPUTED LOAD PARAMETER 1ST, 2ND ORDER RATES.
C
C QQ(I), RRQ(I) = COMPUTED 1ST, 2ND ORDER DISPLACEMENT RATE AT DOF I.
C
C CC(I) = CURRENT CUMULATIVE DISPLACEMENT AT DOF I.

COMMON/COMQQ/QQ
INTEGER NN, KFD(I)
DOUBLE PRECISION RP(1), PRO, PRI, LSIGN, RL, RRL, RQ(1), RRQ(1), QQ(1),
IKMAT(NN, NN), P1(I),
DOUBLE PRECISION RSIGN

INTEGER I

CALL SOLVE(KMAT, NN, KFD, RP, RQ)
CALL PICOMP(NN, QQ, RQ, PRO, PRI, P1)

RL = 0.0
RRL = 0.0
DO 50 I = 1, NN
   IF(KFD(I) .LT. 0) GO TO 49
   RL = RL + RP(I) * RQ(I)
   RRL = RRL + P1(I) * RQ(I)
   GO TO 50
49 P1(I) = 0.0
50 CONTINUE

CALL SOLVE(KMAT, NN, KFD, P1, RRQ)
DO 55 I = 1, NN
   IF(KFD(I) .GT. 0) RL = RL + RP(I) * RRQ(I)
55 CONTINUE

RSIGN = 1.0
IF(RL .LT. 0.0) RSIGN = -1.0

RL = RSIGN / RL

RRL = RSIGN * RL ** 2 * RRL * .50

DO 60 I = 1, NN
60 RRQ(I) = RRL * RQ(I) - RL * RRQ(I)

RL = LSIGN * SQRT(RL)

DO 80 I = 1, NN
80 RC(I) = RL * RQ(I)
RETURN
END

SUBROUTINE STEP(LSIGN, RL, RRL, NN, RQ, RRQ, JUMP, MJUMP, NJUMP, DSLOPE, IPATH, LAHS)
C COMPUTE PATH = PATH DISTANCE, LAMS = LOAD STEP SIZE.
C LSIGN = +,- IF LOAD PARAMETER IS INCREASING, DECREASING.
C RL, RRL = 1ST, 2ND ORDER LOAD PARAMETER RATES.
C NN = TOTAL SYSTEM DOF.
C RQ(I), RRQ(I) = 1ST, 2ND ORDER DIAPLACEMENT RATES AT DCF I.
C JUMP = FRACTION OF LOAD INCREMENT PRECEEDING LIMIT POINT
C AT WHICH LIMIT IS TO BE TRAVERSED.
C MJUMP = MAXIMUM NUMBER OF INCREMENT DIVISIONS TO PERFORM WHEN
C NEARING A LIMIT POINT.
C NJUMP = CURRENT NUMBER OF INCREMENT DIVISIONS TO PERFORM WHEN
C NEARING A LIMIT POINT.
C DSOPE = MAXIMUM SLOPE RATIO (CHANGE/AVERAGE) DURING LOAD STEP.
C PATH = COMPUTED PATH STEP SIZE TO BE TAKEN.
C LAMS = INPUT MAXIMUM, COMPUTED ACTUAL LOAD STEP SIZE.
INTEGER NN, MJUMP, NJUMP
DOUBLE PRECISION LSIGN, RL, RRL, RQ(I), RRQ(I), JUMP, DSOPE, PATH, LAMS
INTEGER N, I
DOUBLE PRECISION SLIM, LAMLIM, PREC, DSOPE, CR, CRR, C

IF(LSIGN*RL.LE.0.00) STOP 601
IF(RL*RRL.GE.0.D0) GO TO 21
POSSIBLE LIMIT POINT
SLIM = -RL/RRL
LAMLIM = DABS(SLIM*RL + .500*SLIM**2*RRL)
CHECK CLOSNESS OF LIMIT POINT
IF(LAMLIM.LT.JUMP) GO TO 11
IF(LAMS.LT.LAMLIM/MJUMP) GO TO 21
LIMIT IS CLOSE. TAKE FRACTIONAL STEP JUMP.
NJUMP = NJUMP-1
IF(NJUMP.LT.2) NJUMP = 2
IF(LAMS.LT.LAMLIM) GO TO 7
STEP IS LARGER THAN LIMIT. JUMP TOWARD LIMIT VALUE.
LAMS = LAMLIM/NJUMP
GO TO 10
STEP IS SMALLER THAN LIMIT. JUMP TOWARD STEP VALUE.
7 N = NJUMP*LAMS/LAMLIM + 1
IF(N.EQ.1) NJUMP = NJUMP+1
LAMS = LAMS/N
10 LAMS = LSIGN*LAMS
PATH = (-RL+LSIGN*DSQRT(RL**2+2.00*LAMS*RRL))/RRL
RETURN
C LIMIT IS VERY CLOSE. TRAVERSE THE LIMIT POINT

11 NJUMP = MJUMP + 1
LSIGN = -LSIGN
LAMS = 0.00
PATH = -2.00*RL/RRL
RETURN

C LIMIT IS NOT CLOSE. CHECK SLOPE CHANGE FOR ALLOWABLE STEP

21 NJUMP = MJUMP + 1
CR = 0.00
CRR = 0.00
DO 50 I = 1, NN
CR = CR + DABS(RQ(I))
50 CRR = CRR + DABS(RRQ(I))
LAMS = LSIGN*LAMS
51 PREC = DABS(2.00*LAMS*RRL/RL**2)
IF(PREC.LT.1.D-8)PATH = LAMS/RL
IF(PREC.GE.1.D-8)PATH = (-RL+LSIGN*CSQRT(RL**2+2.00*LAMS*RRL))/RRL
DSLOP = PATH*RRL
DSLOP = DSLOP/(RL+.50*CSLOP)
C = PATH*CRR
C = C/(CR+.50*C)
IF(DABS(CSLOP).LE.DSLOPE.AND.C.LE.DSLOPE")RETURN
LAMS = .50*LAMS
IF(DABS(LAMS).LE.1.D-3)STOP 602
GC TO 51
END

SUBROUTINE EIGEN1(NN,NCODE,QQDELT,SCRIT,QQPOST,LAM,PRG,PR1,PP)
FORM POSTBUCKLING LOAD TERM PP USING FUNDAMENTAL DISPLACEMENTS
C (REFERENCE VALUES CQSTAR AND CRITICAL INCREMENT QQDELT) AND
C POSTBUCKLING DISPLACEMENT EIGENVECTOR QQPOST.
C NN = TOTAL SYSTEM DOF.
C NCODE = 0, 1 FOR LINEAR, NONLINEAR EIGEN SOLUTION.
C QQDELT(I) = ESTIMATED INCR. DISPLACEMENT AT CUF I TO CRITICAL PT.
C SCRIT = ESTIMATED INCREMENTAL PATH VALUE TO CRITICAL PCINT.
C QQPOST(I) = ESTIMATED BUCKLING DISPLACEMENT AT DOF I.
C LAM = ESTIMATED INCREMENTAL LOAD PARAMETER VALUE TO CRITICAL PT.
C PRG,PR1 = PRESSURE FACTORS AT START, END OF LOAD STEP.
C PP(I) = COMPUTED RIGHT-HAND-SIDE FOR INVERSE POWER ITERATION.
C **ORD** = MAXIMUM TENSOR ORDER TO BE USED FOR DIFFERENCE EXPANSION.
C NEL, NS, NF = NUMBER OF ELEMENTS, STRAIN COMPONENTS, DOF PER NODE.
C ELNO(J,I) = NODE NUMBERS FOR ELEMENT I.
C EGEOM(J,I) = GEOMETRIC PROPERTIES FOR ELEMENT I.
C T(I) = THICKNESS OF ELEMENT I.
C CCSTAR(I) = CURRENT CUMULATIVE DISPLACEMENT AT DOF I.
C IPRESS = NONCONSERVATIVE CODE = 0,1 FOR NC PRESSURE, PRESSURE.
C PRREF(I) = PRESSURE INTENSITY ON ELEMENT I FOR PRESSURE VECTOR.
C COMMON/COMORD/ORD/COMNEL/NEL/COMNS/NS/COMNF/NF
C COMMON/COMEL/ELNO/COMEG/EGEOM/COMT/T/COMQQ/QQSTAR
C COMMON/COMIPR/IPRESS/CMPRESS/PRREF
C INTEGER NN, NCODE, ORD, NEL, NS, NF, ELNC(3,1), IPRESS
C DOUBLE PRECISION QQDELT(l), SCRIT, QPOST(1), LAM, PRO, PRI, PP(1)
C DOUBLE PRECISION EGEOM(3,1), T(l), CCSTAR(1), PRREF(1)
C INTEGER NF2, NF3, IU2, IU3, l, II, M
C DOUBLE PRECISION V, C, QDELT(9), QPOST(9), DSTART(6), DDELT(6)
C 1DPOST(6), ESTAR(3), EDELT(3), EPOST(3), ASTAR(3,6), A1DELT(3,6),
C 2APOST(3,6), STORS(3), STORB(3), STORM(6, G(6,9), P9), USTAR(20), CA
C DOUBLE PRECISION CO, C1, VSTAR(6), VDELT(6), VPOST(6), VI(3), V2(3),
C 1V3(3), V4(3)
C PP(I) = 0.00
C I = 1, NN
C **FORM FUNDAMENTAL REFERENCE QUANTITIES**
C CALL QFILL(NF, ELNO(1,1), QQSTAR, QQSTAR)
C CALL DFILL(NF, EGEOM(1,1), QDELT, QDELT)
C CALL AFILL(NF, DSTART, ASTART, 1)
C CALL EFILL(NF, DSTART, ESTART)
C **FORM STRESS-STRAIN TENSORS**
C CALL UFILL(I, ORD, ESTART, USTART)
C **FORM FUNDAMENTAL CRITICAL INCREMENT QUANTITIES**
C CALL QFILL(NF, ELNO(1,1), QDELT, QDELT)
C CALL DFILL(NF, EGEOM(1,1), QDELT, DDELT)
C CALL AFILL(NF, DDELT, A1DELT, 0)
C DO 20 I = 1, NS
C 5 PP(I) = 0.00
C CO 1CO II = 1, NEL
C V = -5.00*T(II)*EGEOM(1,1)*EGEOM(2,II)
C = 0.00
DO 15 M=1,NF2
CA = ASTAR(I,M)
IF(NCODE.GT.0)CA = CA + .500*SCRIT*AIDELT(I,M)
15 C = C + CA*DDELTA(M)
20 EDEL(T(I)) = C
C FCRM POSTBUCKLING EIGEN QUANTITIES
CALL QFILL(NF,ELNO(I),QPOST,QPOST)
CALL DFILL(NF,EGEOM(I),QPOST,DPOST)
CALL AFILL(NF,DPOST,A1POST,0)
DC 30 I=1,NS
C = 0.00
DO 25 M=1,NF2
CA = ASTAR(I,M)
IF(NCODE.GT.0)CA = CA + SCRIT*AIDELT(I,M)
25 C = C + CA*DPOST(M)
30 EPOST(I) = C
C FORM STORM = DO PORTION OF DISPLACEMENT DERIVATIVE PSUEDO FORCES
DC 104 I=1,NS
C = 0.00
DC 105 M=1,NF2
102 C = C + AIDELT(I,M)*DPCST(M)
104 STORB(I) = C
CALL USUM1(2,USTAR(IU2),NS,STORB,STORA)
DC 110 I=1,NF2
C = 0.00
DO 105 M=1,NS
105 C = C + STORA(M)*ASTAR(M,1)
110 STORM(I) = C
CALL USUM1(2,USTAR(IU2),NS,EPOST,STORA)
DC 120 I=1,NF2
C = 0.00
DO 115 M=1,NS
115 C = C + STORA(M)*AIDELT(M,1)
120 STORM(I) = STORM(I) + C
CALL USUM1(2,USTAR(IU2),NS,EDEL,T,STORA)
DO 130 I=1,NF2
C = 0.00
DO 125 M=1,NS
125 C = C + STORA(M)*A1POST(M,1)

130 STORM(I) = STORM(I) + C  
IF(ORD.LT.3)GO TO 501  
C  
ACD D1 PORTION OF DISPLACEMENT DERIVATIVE PSUEDO FORCES TO STORM  
CALL USUM21(USTAR(IU3),NS,EPOST,EDELT,STORA)  
DO 210 I=1,NF2  
C = 0.DO  
DO 205 M=1,NS  
205 C = C + STORA(M)*ASTAR(M,I)  
210 STORM(I) = STORM(I) + C  
IF(NCODE.EQ.0)GO TO 501  
DO 220 I=1,NF2  
C = 0.DO  
DO 215 M=1,NS  
215 C = C + STORA(M)*AIDELT(M,I)  
220 STORM(I) = STORM(I) + SCRIT*C  
CALL USUM1(3,USTAR(IU3),NS,EDELT,STORA)  
DO 230 I=1,NF2  
C = 0.DO  
DO 225 M=1,NS  
225 C = C + STORA(M)*AIPOST(M,I)  
230 STORM(I) = STORM(I) + .500*SCRIT*C  
C  
COMPUTE ELEMENT FORCES P(I) = VOLUME INTEGRAL OF G(M,I)*STORM(M)  
501 CALL GFILL(NF,EGEOM(1,II),G)  
DO 510 I=1,NF3  
C = 0.DO  
DO 505 M=1,NF2  
505 C = C + G(M,I)*STORM(M)  
510 P(I) = C*V  
IF(IPRESS.EQ.0)GO TO 1000  
C  
ACD NONCONSERVATIVE PRESSURE FORCES TO P  
VSTAR(1) = EGEOM(1,II) + QSTAR(4) - QSTAR(1)  
VSTAR(2) = QSTAR(5) - QSTAR(2)  
VSTAR(3) = QSTAR(6) - QSTAR(3)  
VSTAR(4) = EGEOM(3,II) + QSTAR(7) - QSTAR(1)  
VSTAR(5) = EGEOM(2,II) + QSTAR(8) - QSTAR(2)  
VSTAR(6) = QSTAR(9) - QSTAR(3)  
VCELT(1) = QDELT(4) - QDELT(1)  
VCELT(2) = QDELT(5) - QDELT(2)  
VCELT(3) = QDELT(6) - QDELT(3)  
VDELT(4) = QDELT(7) - QDELT(1)  
00019240  
00019250  
00019260  
00019270  
00019280  
00019290  
00019300  
00019310  
00019320  
00019330  
00019340  
00019350  
00019360  
00019370  
00019380  
00019390  
00019400  
00019410  
00019420  
00019430  
00019440  
00019450  
00019460  
00019470  
00019480  
00019490  
00019500  
00019510  
00019520  
00019530  
00019540  
00019550  
00019560  
00019570  
00019580  
00019590  
00019600  
00019610  
00019620  
00019630
VCELT(5) = QDELT(8) - QDELT(2)
VDELT(6) = QDELT(9) - QDELT(3)
VPOST(1) = QPOST(4) - QPOST(1)
VPOST(2) = QPOST(5) - QPOST(2)
VPOST(3) = QPOST(6) - QPOST(3)
VPOST(4) = QPOST(7) - QPOST(1)
VPOST(5) = QPOST(8) - QPOST(2)
VPOST(6) = QPOST(9) - QPOST(3)
CALL VCRoss(VPOST(4), V1)
CALL VCRoss(VPOST, VSTAR(4), V2)
CALL VCRoss(VDELT, VPOST(4), V3)
CALL VCRoss(VPOST, VDELT(4), V4)
C0 = LAM*(PRI-PRO)*PRREF(II)/6.DO
C1 = PRO*PRREF(II)/6.DO
IF(NCODE.GT.0)C1 = C1 + SCRIT*C0
DO 610 I=1, 3
   C = CO*(V1(I)+V2(I)) + C1*(V3(I)+V4(I))
   P(I) = P(I) - C
   P(I+3) = P(I+3) - C
   P(I+6) = P(I+6) - C
   610
CALL PFILL(NFELNO(I1),P,PP)
RETURN
END

SUBROUTINE EIGEN(UO,KMAT,P1,Q0,NN,KFC,PRO,PRI,RL,RRL,KQ,RRQ, 00019880
   IM1,IM2,ERR1,ERR2,EIG1,Q1,N)
C SOLVE FOR EIGEN LOAD EIG1, AND VECTOR Q1 WITH MAX. INDEX VALUE N.
C UO = OUTPUT UNIT FILE.
C KMAT = SYSTEM JACOBIAN STIFFNESS MATRIX.
C P1 = TEMPORARY FORCE STORAGE VECTOR.
C Q0 = TEMPORARY DISPLACEMENT STORAGE VECTOR.
C NN = TOTAL SYSTEM DOF.
C KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.
C PRO, PRI = PRESSURE FACTORS AT START, END OF LCAD STEP.
C RL, RRL = 1ST, 2ND ORDER LOAD PARAMETER RATES.
C RG(I), RRQ(I) = 1ST, 2ND ORDER DISPLACEMENT RATE AT DOF I.
C M1, M2 = MAXIMUM ITERATIONS FOR LINEAR, LINEAR+NONLINEAR SOLUTION.
C ERR1, ERR2 = MAXIMUM ERROR FOR LINEAR, LINEAR+NONLINEAR SOLUTION.
C EIG1 = COMPUTED EIGENVALUE (CRITICAL PATH VALUE).
C Q1(I) = I TH COMPONENT OF EIGENVECTOR (BUCKLING DISPLACEMENT).
C N = DOF OF LARGEST Q1 COMPONENT.
COMMON/COMNF/NF
INTEGER UO, NN, KFD(I), MI1, MI2, N
DOUBLE PRECISION KMAT(NN, NN), P1(I), QO(I), PRO, PR1,
RRL, RRL, RRQ(I), ERR1, ERR2, EIG1, Q1(I)
INTEGER I, NI
DOUBLE PRECISION EIG0, MAX1, DL, C

201 FORMAT(1H 'EIGENVALUE = ', D15.8, 5X, 'DOF OF LARGEST COMPONENT OF EIGENVECTOR = ', 15)
N1 = 0
NCODE = 0
EIG1 = 0. DO
DO 5 I=1, NN
5 Q1(I) = 1
11 EIGO = EIG1
   DL = RL
   IF(NCODE.GT.0) DL = DL + .5D0*EIG0*RRL
   DO 20 I=1, NN
   QO(I) = Q1(I)
   Q1(I) = RRQ(I)
   IF(NCODE.GT.0) Q1(I) = Q1(I) + .5D0*EIG0*RRQ(I)
20 CONTINUE
   CALL EIGEN1(NN, NCDE, Q1, EIGO, QQ, CL, PRO, PR1, P1)
   NOD = NN/NF
   CALL OUTPQ(6, NOD, NF, P1, QO)
   DO 50 I=1, NN
   IF(KFD(I).LT.0) PI(I) = 0.D0
50 CONTINUE
   CALL SOLVE(KMAT, NN, KFD, P1, Q1)
   N = 0
   MAX1 = 0. DO
   DO 70 I=1, NN
   IF(DABS(Q1(I)).LE.DABS(MAX1)) GO TO 70
   N = I
   MAX1 = Q1(I)
70 CONTINUE
   MAX1 = 1.D0/MAX1
   EIG1 = -MAX1
IF(KFD(N).NE.N) N = KFD(N)
DC 100 I=1,NN
100 Q1(I) = Q1(I)*MAX1
WRITE(UO,2011)EIG1,N
NI = NI+1
C = DABS((EIG1-EIG0)/EIG1)
IF(NCODE.GT.O.AND.(NI.GE.MI2.OR.C.LE.ERR2)) RETURN
IF(NCODE.EQ.0.AND.(NI.GE.MI1.OR.C.LE.ERR1)) NCODE = 1
GO TO 11
END

SUBROUTINE POST2(NN, QQCRIT, QQDCT, QQPOST, LCRIT, LDOT, PRO, PRI, PP, IKODE)

C FORM 2ND ORDER POSTBUCKLING LOAD TERM PP (=P21, P22 FOR KODE=1,2).

C NN = TOTAL SYSTEM DOF.

C QQCRIT(I) = PREDICTED TOTAL DISPLACEMENT OF DOF I AT CRITICAL PT.

C QQDCT(I) = PREDICTED DISPLACEMENT RATE OF DOF I AT CRITICAL PT.

C QQPOST(I) = ITH DISPLACEMENT OF CRITICAL BUCKLING EIGENVECTOR.

C LCRIT = PREDICTED LOAD PARAMETER VALUE AT CRITICAL POINT.

C LDOT = PREDICTED LOAD PARAMETER RATE AT CRITICAL POINT.

C PRO, PRI = PRESSURE FACTORS AT START OF LOAD STEP, CRITICAL POINT.

C PP(I) = COMPUTED POSTBuckling FORCE TERM AT DOF I.

C KODE = CODE FOR DUAL USE OF SUBROUTINE.

C NEL, NS, NF = NUMBER OF ELEMENTS, STRAIN COMPONENTS, DOF PER NODE.

C EGEOM(J, I) = GEOMETRIC PROPERTIES FOR ELEMENT I.

C T(I) = THICKNESS OF ELEMENT I.

C IPRESS = NONCONSERVATIVE CODE = 0, 1 FOR NO PRESSURE, PRESSURE.

C PRREF(I) = PRESSURE INTENSITY ON ELEMENT I FOR PRESSURE VECTOR.

COMMON/COMORD/ORD/COMNEL/NS/COMNS/NF/COMNF/NF
COMMON/COMEL/ELNO/COMEG/EGEOM/COMT/T
COMMON/CCMORD/IPRESS/CCMPRR/PRREF
INTEGER NN, KODE, ORD, NEL, NS, NF, ELNO(3,1), IPRESS
DOUBLE PRECISION QQCRIT(1), QQDCT(1), QQPOST(1), LCRIT, LDOT, PRO, PRI,
PP(1), EGEOM(3,1), T(1), PRREF(1)
INTEGER NF2, NF3, IU2, IU3, I, J, M
DOUBLE PRECISION V, C, QQCRIT(9), QQDCT(9), QQPOST(9), LCRIT(6), LDOT(6),
PP(6), ECRIT(3), EDOT(3), EPOST(3), ACRIT(3,6), AIDOT(3,6),
1AIPCST(3,6),STORA(3),STORH(3),STORM(6),G(6,9),P(9),UCRIT(20)
CCUBE PRECISION C,C1,VCRT(6),VDCT(6),VPOST(6),V1(3),V2(3),
IV3(3),V4(3)

NF2 = NF*2
NF3 = NF*3
IU2 = 2 + NS
IU3 = IU2 + NS*(NS+1)/2
DO 5 I=1,NN
5 PP(I) = 0.0
DO 1000 II=1,NEL
V = 0.5D0*T(II)*EGECM(1,II)*EGECM(2,II)
C FCRM FUNCTIONAL CRITICAL QUANTITIES
CALL QFILL(NF,ELNC(1,II),QCRIT,QCRIT)
1901 FORMAT(1HO,9E14.7)
WRITE(6,1901)(QCRIT(I),I=1,NF3)
CALL DFILL(NF,EGECM(1,II),QCRIT,DCRIT)
WRITE(6,1901)(DCRIT(I),I=1,NF2)
CALL AFILL(NF,0CRIT,ACRIT,1)
DO 1910 I=1,3
1910 WRITE(6,1901)(ACRIT(I,J),J=1,NF2)
CALL EFILL(NF,QCRIT,ECRIT)
WRITE(6,1901)(ECRIT(I),I=1,3)
C FORM STRESS-STRAIN TENSORS
C FCRM FUNCTIONAL RATE QUANTITIES
CALL QFILL(NF,ELNC(1,II),QDOT,QDOT)
WRITE(6,1901)(QDOT(I),I=1,NF3)
CALL DFILL(NF,EGECM(1,II),0COT,DCOT)
WRITE(6,1901)(DCOT(I),I=1,NF2)
CALL AFILL(NF,0COT,ADOT,0)
DO 1920 I=1,3
1920 WRITE(6,1901)(ADOT(I,J),J=1,NF2)
DO 20 I=1,NS
C = 0.0
DO 15 M=1,NF2
15 C = C + ACRIT(I,M)*DDCT(M)
20 EDOT(II) = C
WRITE(6,1901)(EDOT(I),I=1,3)
C FCRM POSTBUCKLING EIGEN QUANTITIES
CALL QFILL(NF,ELNC(1,II),QPOST,QPOST)
WRITE(6,1901)(QPCST(I),1=1,NF3) 00021200
CALL DFILL(NF,EGEOM(I,II),QPCST,DPCST) 00021210
WRITE(6,1901)(DPOST(I),1=1,NF2) 00021220
CALL AFILL(NF,DPOST,A1POST,O) 00021230
CC 1930 I=1,3 00021240
1930 WRITE(6,1901)(A1POST(I,J),J=1,NF2) 00021250
DO 30 I=1,NS 00021260
C = 0.00 00021270
DO 25 M=1,NF2 00021280
25 C = C + ACRIT(I,M)*DPOST(M) 00021290
30 EPOST(I) = C 00021300
WRITE(6,1901)(EPOST(I),I=1,3) 00021310
C FORM STORM = DO PORTION OF DISPLACEMENT DERIVATIVE PSUEDO FORCES 00021320
DO 104 I=1,NS 00021330
C = 0.00 00021340
DO 102 M=1,NF2 00021350
102 C = C + A100T(I,M)*DPOST(M) 00021360
104 STORB(I) = C 00021370
CALL USUM12,UCRIT(IU2),NS,STORB,STORA) 00021380
DO 110 I=1,NF2 00021390
C = 0.00 00021400
DO 105 M=1,NS 00021410
105 C = C + STORA(M)*ACRIT(M,I) 00021420
110 STORM(I) = C 00021430
WRITE(6,1901)(STORM(I),I=1,NF2) 00021440
CALL USUM12,UCRIT(IU2),NS,EDCT,STORA) 00021450
DO 120 I=1,NF2 00021460
C = 0.00 00021470
DO 115 M=1,NS 00021480
115 C = C + STORA(M)*A1POST(M,I) 00021490
120 STORM(I) = STORM(I) + C 00021500
WRITE(6,1901)(STORM(I),I=1,NF2) 00021510
CALL USUM12,UCRIT(IU2),NS,EPOST,STORA) 00021520
DO 130 I=1,NF2 00021530
C = 0.00 00021540
DO 125 M=1,NS 00021550
125 C = C + STORA(M)*A1DCT(M,I) 00021560
130 STORM(I) = STORM(I) + C 00021570
WRITE(6,1901)(STORM(I),I=1,NF2) 00021580
IF(ORD.LT.3)GO TO 501 00021590
ACD D1 PORTION OF DISPLACEMENT DERIVATIVE PSUEDO FORCES TO STORM
CALL USUM21(UCRIT(IU3),NS,EDCT,EPOST,STORA)
DO 210 I=1,NF2
  C = 0.DO
  DO 205 M=1,NS
  205 C = C + STORA(M)*ACRIT(M,I)
  210 STORM(I) = STORM(I) + C
CALL COMPUTE ELEMENT FORCES P(I) = VOLUME INTEGRAL OF G(M,I)*STORM(M)
DO 510 I=1,NF3
  C = 0.DO
  DO 505 M=1,NF2
  505 C = C + G(M,I)*STORM(M)
  510 P(I) = C*V
WRITE(6,1901)(P(II,I=,NF3)
IF(IPRESS.EQ.0)GO TO 1000
ADD NONCONSERVATIVE PRESSURE FORCES TO P
VPDCT(1) = QDOT(4) - QDCT(1)
VPDCT(2) = QDOT(5) - QDCT(2)
VPDCT(3) = QDOT(6) - QDCT(3)
VPDCT(4) = QDOT(7) - QDCT(1)
VPDCT(5) = QDOT(8) - QDCT(2)
VPDCT(6) = QDOT(9) - QDCT(3)
VPOST(1) = QPOST(4) - QPOST(1)
VPOST(2) = QPOST(5) - QPOST(2)
VPOST(3) = QPOST(6) - QPOST(3)
VPOST(4) = QPOST(7) - QPOST(1)
VPOST(5) = QPOST(8) - QPOST(2)
VPOST(6) = QPOST(9) - QPOST(3)
C0 = (PRO + LCRIT*(PR1-PRO))*PRREF(I1)/6.DO
C1 = 0.DO
CALL VCROSS(VDCT,VPOST(4),V1)
CALL VCROSS(VPOST,VDCT(4),V2)
IF(KODE.EQ.2)GO TO 601
VCRIT(1) = EGEOM(1,II) + QCRIT(4) - QCRIT(1)
VCRIT(2) = QCRIT(5) - QCRIT(2)
VCRIT(3) = QCRIT(6) - QCRIT(3)
VCRIT(4) = EGEOM(3,II) + QCRIT(7) - QCRIT(1)
VCRIT(5) = EGEOM(2,II) + QCRIT(8) - QCRIT(2)
VCRIT(6) = QCRIT(9) - QCRIT(3)
C1 = LDTO*(PR1-PR)*PRREF(II)/6.00
CALL VCROSS(VCRIT, VPOST(4), V3)
CALL VCROSS(VPOST, VCRIT(4), V4)
601 DO 610 I=1,3
C = CO*(V1(I)+V2(I)) + C1*(V3(I)+V4(I))
P(I) = P(I) - C
P(I+3) = P(I+3) - C
610 P(I+6) = P(I+6) - C
ACC ELEMENT FORCES P TO SYSTEM FORCES PP
1000 CALL PFILL(NF,ELNC(1,II), P, PP)
RETURN
END

SUBROUTINE POST3(NN, QQCRIT, QQDGT1, QQPOS1, QQDCT2, QQPOS2, SPOST1,
1 LCRIT, LDOT1, LDOT2, PR1, PP)
C FORM 3RD ORDER POSTBUCKLING LOAD TERM PP.
C NN = TOTAL SYSTEM DOF.
C QQCRIT(I) = PREDICTED TOTAL DISPLACEMENT OF COF I AT CRITICAL PT.
C QQDGT1(I) = CRITICAL 1ST ORDER DISPLACEMENT RATE AT COF I.
C QQPOS1(I) = ITH EIGENVECTOR VALUE (1ST ORDER POSTBUCKLING RATE).
C QQDCT2(I) = CRITICAL 2ND ORDER DISPLACEMENT RATE AT COF I.
C QQPOS2(I) = ITH VALUE OF 2ND ORDER POSTBUCKLING DISPLACEMENT.
C SPOST1 = 1ST ORDER FUNDAMENTAL PATH RATE.
C LCRIT = PREDICTED LOAD PARAMETER VALUE AT CRITICAL POINT.
C LDOT1 = 1ST ORDER LOAD PARAMETER RATE AT CRITICAL POINT.
C LDOT2 = 2ND ORDER LOAD PARAMETER RATE AT CRITICAL POINT.
C PR1 = PRESSURE FACTORS AT START OF LOAD STEP, CRITICAL POINT.
C PP(I) = COMPUTED PSUEDO FORCE TERM AT DOF I.
C ORD = MAXIMUM TENSOR ORDER TO BE USED FOR DIFFERENCE EXPANSION.
C NEL, NS, NF = NUMBER OF ELEMENTS, STRAIN COMPONENTS, DOF PER NODE.
C ELNO(J, I) = NODE NUMBERS FOR ELEMENT I.
C EGEOM(J, I) = GEOMETRIC PROPERTIES FOR ELEMENT I.
C T(I) = THICKNESS OF ELEMENT I.
C IPRESS = NONCONSERVATIVE CODE = 0,1 FOR NO PRESSURE, PRESSURE.
C PRCF(I) = PRESSURE INTENSITY ON ELEMENT I FOR PRESSURE VECTOR.
COMMON/COMND,CN/COMOR,CORD/COMNEL/NEL/COMNS/NS/COMNF/NF
COMMON/COMOL/ELNC/CNEM/EGEOM/CINT/T
COMMON/COMIPR/IPRESS/CCMPPR/PRREF
INTEGER NN, ORD, NEL, NS, NF, ELNO(3, I), IPRESS
DOUBLE PRECISION C, CRIT(1), QQDOT1(1), QQPCST1(1), QQDOT2(1),
1QQPOST2(1), SPOST1, LCRIT, LDOT1, LDOT2, PRO, P1, PP(1), EGEOM(3,1), T(1),
2PRREF(1)

INTEGER NF2, NF3, IU2, IU3, I, II, M

DOUBLE PRECISION V, C, CRIT(9), QQDOT1(9), QQPOST1(9), QQDOT2(9),
1QQPOST2(9), OCRIT(6), DDOT1(6), DPOST1(6), DPOST2(6),
2ECRIT(3), EPOST1(3), EPOST2(3), ACRIT(I,6),
3A1DOT1(3,6), A1POST1(3,6), A1POST2(3,6), STORA(3), STORB(3),
4STORM(6), G(6,9), P(9), UCRT1(20)

DOUBLE PRECISION C, C1, C2, VCRIT(6), VDOT1(6), VPOST1(6), VDOT2(6),
2V10(3), V11(3), V12(3), V13(3)

NF2 = NF#2
NF3 = NF#3
IU2 = 2 + NS
IU3 = IU2 + NS*(NS+1)/2

DO 5 I=1, N

PP(I) = 0.D0

DO 1000 II=1, NEL

V = .5*DO*(II)*EGECM(1, II)*EGECM(2, II)

C FCRM FUNDAMENTAL CRITICAL QUANTITIES

CALL QFILL(NF, ELOO(I, II), QQCRIT, QCRT)

CALL DFILL(NF, EGEOM(1, II), QCRT, QCRT)

CALL AFILL(NF, QCRT, ACRIT, 1)

CALL EFILL(NF, QCRT, ECRIT)

C FCRM STRESS-STRAIN TENSORS

CALL UFILL(II, CRC, ECRIT, UCRT1)

C FCRM 1ST ORDER FUNDAMENTAL RATE QUANTITIES

CALL QFILL(NF, ELOO(I, II), QQDOT1, QQDOT1)

CALL DFILL(NF, EGEOM(1, II), QQDOT1, QQDOT1)

CALL AFILL(NF, QQDOT1, A1DOT1, 0)

DO 20 I=1, NS

C = 0.D0

DO 15 M=1, NF2

C = C + ACRIT(I, M)*ODOT1(M)

20 EDOT1(I) = C

C FCRM 1ST ORDER POSTBUCKLING RATE QUANTITIES

CALL QFILL(NF, ELOO(I, II), QQPOST1, QQPOST1)

CALL DFILL(NF, EGEOM(1, II), QQPOST1, QQPOST1)

CALL AFILL(NF, QQPOST1, A1POST1, 0)
DO 30 I=1,NS
C = 0.00
DO 25 M=1,NF
25 C = C + ACRIT(I,M)*DPOST1(M)
30 EPOST1(I) = C
C FORM 2ND ORDER FUNDAMENTAL RATE QUANTITIES
CALL QFILL(NF,ELNC(1,II),QQCTS2,QDCTS)
CALL DFILL(NF,EGRM(1,II),QDOTS2,CDCTS2)
CALL AFILL(NF,DDOTS2,A1DOTS2,0)
DC 40 I=1,NS
C = 0.00
DO 35 M=1,NF
35 C = C + ACRIT(I,M)*DDOTS2(M) + A1DOT1(I,M)*CDCTS1(M)
40 EDOT2(I) = C
DO 40 E stor = 2ND ORDER POSTBUCKLING RATE QUANTITIES
CALL QFILL(NF,ELNC(1,II),QQPCS2,QPOST2)
CALL DFILL(NF,EGRM(1,II),QPOST2,QPOST2)
CALL AFILL(NF,DPOST2,A1POS2,0)
DO 50 I=1,NS
C = 0.00
DO 45 M=1,NF
45 C = C + ACRIT(I,M)*DPOST2(M) + A1POS1(I,M)*DPOST1(M) + 2.00*
1DPOST1*CDCTS1(M))
50 EPOST2(I) = C
C FORM STORM = DO PORTION OF DISPLACEMENT DERIVATIVE PSUEDO FORCES
DO 104 I=1,NS
C = 0.00
DO 102 M=1,NF
102 C = C + A1POS1(I,M)*CDCTS2(M)
104 STORB(I) = C
CALL USUMI(2,UCRIT(IU2),NS,STORB,STORA)
DO 110 I=1,NS
C = 0.00
DC 105 M=1,NS
105 C = C + STOR(A*M) + ACRIT(M,I)
110 STORM(I) = SPOST1**2*C
CALL USUMI(2,UCRIT(IU2),NS,STOR1,STORA)
DO 120 I=1,NS
C = 0.00
DO 115 M=1,NS
C = C + STORA(M)*A1DOT2(M,I)

STORM(I) = STORM(I) + SPOST1**2*C
CALL USUM1(2,UCRIT(IU2),NS,EDOT2,STORA)
DC 130 I=1,NF2
C = 0.0D0
DO 125 M=1,NS
125 C = C + STORA(M)*A1POS1(M,I)
STORM(I) = STORM(I) + SPOST1**2*C
DO 130 I=1,NS
C = 0.0D0
DO 132 M=1,NS
132 C = C + STORM(I) + SPOST1*C
CALL USUM1(2,UCRIT(IU2),NS,STORA)
DC 140 I=1,NS
C = 0.0D0
DO 135 M=1,NS
135 C = C + STORA(M)*ACRIT(M,I)
STORM(I) = STORM(I) + SPOST1*C
CALL USUM1(2,UCRIT(IU2),NS,EDOT1,STORA)
DC 150 I=1,NS
C = 0.0D0
DO 145 M=1,NS
145 C = C + STORA(M)*A1POS2(M,I)
STORM(I) = STORM(I) + SPOST1*C
CALL USUM1(2,UCRIT(IU2),NS,EPOST2,STORA)
DC 160 I=1,NS
C = 0.0D0
DO 155 M=1,NS
155 C = C + STORA(M)*A1DOT1(M,I)
STORM(I) = STORM(I) + SPOST1*C
DO 160 I=1,NS
C = 0.0D0
DO 162 M=1,NS
162 C = C + STORM(I) + SPOST1*C
STORM(I) = STORM(I) + SPOST1*C
CALL USUM1(2,UCRIT(IU2),NS,STORA)
DC 170 I=1,NS
C = 0.0D0
DO 165 M=1,NS
165 C = C + STORA(M) * ACRIT(M*I)
170 STORM(I) = STORM(I) + C
    CALL USUM1(2, UCRIT(IU2), NS, EPOST1, STCRA)
    DO 180 I=1, NF2
    C = 0.00
    DO 175 M=1, NS
    175 C = C + STORA(M) * APOS2(M, I)
180 STORM(I) = STORM(I) + C
    CALL USUM1(2, UCRIT(IU2), NS, EPOST2, STCRA)
    DO 190 I=1, NF2
    C = 0.00
    DO 185 M=1, NS
    185 C = C + STORA(M) * APOS1(M, I)
190 STORM(I) = STORM(I) + C
    IF (ORD.LT.3) GO TO 501

C ACD C1 PORTION OF DISPLACEMENT DERIVATIVE PSUEDO FORCES TO STORM
200 CALL USUM21(UCRIT(IU3), NS, EPOST1, EDOT2, STCRA)
    DO 210 I=1, NF2
    C = 0.00
    DO 205 M=1, NS
    205 C = C + STORA(M) * ACRIT(M, I)
210 STORM(I) = STORM(I) + SPOST1**2*C
    CALL USUM1(3, UCRIT(IU3), NS, EDOT1, STCRA)
    DO 220 I=1, NF2
    C = 0.00
    DO 215 M=1, NS
    215 C = C + STORA(M) * APOS1(M, I)
220 STORM(I) = STORM(I) + SPOST1**2*C
    CALL USUM21(UCRIT(IU3), NS, EPOST1, EDOT1, STCRA)
    DO 230 I=1, NF2
    C = 0.00
    DO 225 M=1, NS
    225 C = C + STORA(M) * ADOT1(M, I)
230 STORM(I) = STORM(I) + 2.00*SPOST1**2*C
    CALL USUM21(UCRIT(IU3), NS, EDOT1, EPOST2, STCRA)
    DO 240 I=1, NF2
    C = 0.00
    DO 235 M=1, NS
    235 C = C + STORA(M) * ACRIT(M, I)
240 STORM(I) = STORM(I) + SPOST1*C
CALL USUMI(3, UCRIT(IU3), NS, EPOST1, STORA)
DO 250 I=1, NF2
C = 0.00
DC 245 M=1, NS
245 C = C + STORA(M) * AIPOS1(M, I)
250 STORM(I) = STORM(I) + SPOST1*C
CALL USUM21(UCRIT(IU3), NS, EDCIT1, EPCST1, STORA)
DO 260 I=1, NF2
C = 0.00
DO 255 M=1, NS
255 C = C + STORA(M) * AIPOS1(M, I)
260 STORM(I) = STORM(I) + 2.00 * SPOST1*C
CALL USUM21(UCRIT(IU3), NS, EPCST1, EPOST2, STORA)
DC 270 I=1, NF2
C = 0.00
DO 265 M=1, NS
265 C = C + STORA(M) * AIPOS1(M, I)
270 STORM(I) = STORM(I) + C
CALL USUMI(3, UCRIT(IU3), NS, EPOST1, STORA)
DO 280 I=1, NF2
C = 0.00
DO 275 M=1, NS
275 C = C + STORA(M) * AIPOS1(M, I)
280 STORM(I) = STORM(I) + C
C COMPUTE ELEMENT FORCES P(I) = VOLUME INTEGRAL OF G(M, I)*STORM(M)
501 CALL GFILL(NF, EGEOM(1, II), G)
DC 510 I=1, NF3
C = 0.00
DO 505 M=1, NF2
505 C = C + G(M, I) * STORM(M)
510 P(I) = C*V
IF(IPRESS.EQ.0)GO TO 1000
C ADD NONCONSERVATIVE PRESSURE FORCES TO P
VCRT(1) = EGEOM(1, II) + QCRT(4) - QCRT(1)
VCRT(2) = QCRT(5) - QCRT(2)
VCRT(3) = QCRT(6) - QCRT(3)
VCRT(4) = EGEOM(3, II) + QCRT(7) - QCRT(1)
VCRT(5) = EGEOM(2, II) + QCRT(8) - QCRT(2)
VCRT(6) = QCRT(9) - QCRT(3)
VCRT(11) = QDCT1(4) - QCRT1(1)
VCOT1(2) = QDOT1(5) - QCOT1(2)
VCOT1(3) = QDOT1(6) - QCOT1(3)
VCOT1(4) = QDOT1(7) - QCOT1(1)
VCOT1(5) = QDOT1(8) - QCOT1(2)
VCOT1(6) = QDOT1(9) - QCOT1(3)
VPOST1(1) = QPOST1(4) - QPOST1(1)
VPOST1(2) = QPOST1(5) - QPOST1(2)
VPOST1(3) = QPOST1(6) - QPOST1(3)
VPOST1(4) = QPOST1(7) - QPOST1(1)
VPOST1(5) = QPOST1(8) - QPOST1(2)
VPOST1(6) = QPOST1(9) - QPOST1(3)
VDOT2(1) = QDOT1(4) - QCOT1(1)
VDOT2(2) = QDOT1(5) - QCOT1(2)
VDOT2(3) = QDOT1(6) - QCOT1(3)
VDOT2(4) = QDOT1(7) - QCOT1(1)
VDOT2(5) = QDOT1(8) - QCOT1(2)
VDOT2(6) = QDOT1(9) - QCOT1(3)
VPOST2(1) = QPOST2(4) - QPOST2(1)
VPOST2(2) = QPOST2(5) - QPOST2(2)
VPOST2(3) = QPOST2(6) - QPOST2(3)
VPOST2(4) = QPOST2(7) - QPOST2(1)
VPOST2(5) = QPOST2(8) - QPOST2(2)
VPOST2(6) = QPOST2(9) - QPOST2(3)
C0 = (PRO + LCRIT*(PR1-PR0)*PRREF(11))/6.00
C1 = LDOT1*(PR1-PR0)*PRREF(11)/6.00
C2 = LDOT2*(PR1-PR0)*PRREF(11)/6.00
CALL VCROSS(VDOT2, VPOST1(4), V1)
CALL VCRCSS(VPOST1, VDOT2(4), V2)
CALL VCRSS(VDOT1, VPOST1(4), V3)
CALL VCRSS(VPOST1, VDOT1(4), V4)
CALL VCRSS(VCRIT, VPOST1(4), V5)
CALL VCRSS(VPOST1, VCRIT(4), V6)
CALL VCRSS(VDOT1, VPOST2(4), V7)
CALL VCRSS(VPOST2, VDOT1(4), V8)
CALL VCRSS(VCRIT, VPOST2(4), V9)
CALL VCRSS(VPCST1, VCRIT(4), V10)
CALL VCRSS(VPOST1, VPOST1(4), V11)
CALL VCRSS(VPOST2, VPOST1(4), V12)
CALL VCRSS(VPOST1, VPOST2(4), V13)
DO 610 I=1,3
END
C = CO*(V1(I)+V2(I)+V7(I)+V8(I)+V12(I)+V13(I)) + C1*(2.DO*V3(I) + 1.02.DO*V4(I)+V9(I)+V10(I)+2.0*V11(I)) + C2*(V5(I)+V6(I))

P(I) = P(I) - C
P(I+3) = P(I+3) - C
P(I+6) = P(I+6) - C

ADD ELEMENT FORCES P TO SYSTEM FORCES PP

CALL PFILL(NF,ELNO(1,II),P,PP)
RETURN

END

SUBROUTINE PRATES(KMAT,P11,P2,Q2,NN,IPOST,KFD,SCRIT,
QQCRIT,QQDOT1,QQDOT2,LCRIT,LDOT1,LDOT2,PRO,PR1,
2CQPOS1,LPOST1,CQPOS2,LPOST2,CQPOS3)

C COMPUTE POSTBUCKLING RATES FOR LOAD PARAMETER AND DISPLACEMENTS.
C KMAT = SYSTEM JACOBIAN STIFFNESS MATRIX.
C P1, Q1, P2, Q2 = TEMPORARY STORAGE VECTORS.
C NN = TOTAL SYSTEM DOF.
C IPOST = DOF OF LARGEST EIGENVECTOR COMPONENT.
C KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.
C SCRIT = PREDICTED CRITICAL VALUE OF FUNDAMENTAL PATH.
C QQCRIT(I) = PREDICTED CRITICAL VALUE OF DISPLACEMENT AT DOF I.
C QQDOT1(I), QQDOT2(I) = 1ST, 2ND ORDER CRITICAL FUNDAMENTAL DISPLACEMENT RATES AT COF I.
C LCRIT = PREDICTED CRITICAL VALUE OF LOAD PARAMETER.
C LDOT1, LDOT2 = 1ST, 2ND ORDER CRITICAL LOAD PARAMETER RATES.
C PRO, PR1 = PRESSURE FACTORS AT START OF LOAD STEP, CRITICAL POINT.
C LPOST1, LPOST2 = COMPUTE 1ST, 2ND ORDER POSTBUCKLING LOAD RATES.
C CQPOS1(I), CQPOS2(I), CQPOS3(I) = 1ST, 2ND, 3RD ORDER POSTBUCKLING DISPLACEMENT RATES AT DOF I.
C NF = DOF PER NODE.
C COMMON/COMNF/NF

INTEGER NN, IPOST, KFD(I)
DOUBLE PRECISION KMAT(NN,NN), P1(I), Q1(I), P2(I), Q2(I), SCRIT,
1QQCRIT(I), QQDOT1(I), QQDOT2(I), LCRIT, LDOT1, LDOT2, PRO, PR1,
2CQPOS1(I), LPOST1, CQPOS2(I), LPOST2, CQPOS3(I)

INTEGER I
DOUBLE PRECISION CN, CO, SPST1, SPST2

C SOLVE FOR 1ST ORDER DISPLACEMENTS
DO 10 I=1, NN
10 PI(I) = 0.DO
PI(IPOST) = 1.DO
CALL SOLVE(KMAT,NN,KFD,P1,QQPOS1)
NCC = NN/NF
CALL OUTPQ(6,NOD,NF,P1,QQPOS1)
C SOLVE FOR 1ST ORDER LOADS AND 2ND ORDER DISPLACEMENTS
CALL POST2(NN,QQCrit,CQDOT1,QQPOS1,LCRIT,LDOT1,PRO,PR1,P1,1)
CALL POST2(NN,QQCrit,QQPOS1,QQPOS1,LCRIT,LDOT1,PRO,PR1,P1,2)
CALL OUTPQ(6,NOD,NF,P1,P2)
CN = 0.DO
CC = 0.DO
DC 110 I=1,NN
CN = CN + QQPOS1(I)*P2(I)
110 CC = CD + QQPOS1(I)*PI(I)
SPOST1 = -.5DO*CN/CD
LPOST1 = LDOT1*SPOST1
DC 120 I=1,NN
IF(KF(I).GT.0)GO TO 120
PI(I) = 0.DO
P2(I) = 0.DO
CONTINUE
120 CALL SOLVE(KMAT,NN,KFD,P1,Q1)
CALL SOLVE(KMAT,NN,KFD,P2,Q2)
CALL OUTPQ(6,NOD,NF,P1,P2)
CALL OUTPQ(6,NOD,NF,P1,P2)
DO 130 I=1,NN
130 QQPOS2(I) = -(2.DO*SPOST1*Q1(I) + Q2(I))
CALL OUTPQ(6,NOD,NF,QQDOT2,QQPOS2)
C SOLVE FOR 2ND ORDER LOADS AND 3RD ORDER DISPLACEMENTS
CALL POST3(NN,QQCrit,CQDOT1,QQPOS1,QQDOT2,QQPOS2,SPOST1,
LCRIT,LDOT1,LDOT2,PRO,PR1,P2)
CALL OUTPQ(6,NOD,NF,QQCRIT,P2)
CALL OUTPQ(6,NOD,NF,QQDOT1,QQPOS1)
CN = 0.DO
DO 210 I=1,NN
210 QQPOS2(I) = -(QQPOS1(I)*P2(I))
SPOST2 = -CN/CD
LPOST2 = LDOT1*SPOST1 + LDOT2*SPOST2
DO 220 I=1,NN
IF(KF(I).GT.0)GO TO 220
P2(I) = 0.00

CONTINUE

CALL SOLVE(KMAT,NN,KFD,P2,Q2)
CALL OUTPQ(6,NOD,NF,P2,Q2)
DO 230 I=1,NN

CALL SOLVE(KMAT,NN,KFD,P2,Q2)
CALL OUTPQ(6,NOD,NF,QQPOS3,QQPOS3)
RETURN

DEBUG INIT(SPOST1,LPOST1,SPOST2,LPCST2,CN,CD)
END

DOUBLE PRECISION FUNCTION VDOT(V1,V2)
COMPUTE VECTOR DOT PRODUCT VDOT = V1 DOT V2.
DOUBLE PRECISION V1(1),V2(1)
VDOT = V1(1)*V2(1) + V1(2)*V2(2) + V1(3)*V2(3)
RETURN
END

SUBROUTINE VCROSS(V1,V2,VV)
COMPUTE VECTOR CROSS PRODUCT V1 X V2 = VV.
DOUBLE PRECISION V1(1),V2(1),VV(1)
VV(1) = V1(2)*V2(3) - V1(3)*V2(2)
VV(2) = V1(3)*V2(1) - V1(1)*V2(3)
VV(3) = V1(1)*V2(2) - V1(2)*V2(1)
RETURN
END

DOUBLE PRECISION FUNCTION VLENTH(V)
COMPUTE VLENTH = LENGTH OF VECTOR V.
DOUBLE PRECISION V(1)
VLENTH = DSQRT(V(1)**2 + V(2)**2 + V(3)**2)
RETURN
END

SUBROUTINE VNORM(V,VV)
COMPUTE NORMALIZED UNIT VECTOR VV FROM GIVEN VECTOR V.
DOUBLE PRECISION V(1),VV(1),C
C = DSQRT(V(1)**2 + V(2)**2 + V(3)**2)
IF(C.LE.0.DO)C = 1.DO
VV(1) = V(1)/C
VV(2) = V(2)/C
VV(3) = V(3)/C
RETURN
END

SUBROUTINE MERGE(KMAT,ELNO,KFD,NEL,NN,NF)
C GENERATE AND MERGE ELEMENTAL MATRICES INTO GLOBAL MATRIX KMAT.
C ASSUMES ELEMENTS HAVE 3 NODES, MATRICES MAY BE UNSYMMETRIC.
C KMAT = GLOBAL (TOTAL SYSTEM) MATRIX, WHICH MAY BE UNSYMMETRIC.
C ELNO(J,I) = NODE NUMBERS FOR ELEMENT I.
C KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.
C KFD(I) = +,- = SPECIFIED FORCE,DISPLACEMENT.
C KFD(I) = ++,- I INDICATES INDEPENDENT DOF.
C KFD(I) = ++,J INDICATES DEPENDENT FREEDOM WHOSE DISPLACEMENT IS
C CONstrained TO EQUAL DISPLACEMENT AT J.
C NEL = NUMBER OF ELEMENTS.
C NN = TOTAL SYSTEM DOF = NUMBER OF NODES TIMES NF.
C NF = DOF PER NODE.
C GENER8 ROUTINE GENERATES ELEMENTAL MATRIX.
INTEGER ELNO(3,1),KFD(1),NEL,NN,NF
DOUBLE PRECISION KMAT(NN,NN)
INTEGER IE,1,1,II,JI,IO,JG,IO,JJC,IP,JP,IN,JN,ILOC,JLOC
DOUBLE PRECISION K(9,9)
C ZERO OUT GLOBAL MATRIX
DO 5 II=1,NN
DO 5 JJ=1,NN
5 KMAT(II,JI) = 0.DO
DO 100 IE=1,NEL
CALL GENER8(IE,K)
C IP,JP ARE PARTITION ROW,COLUMN NUMBERS.
DO 100 IP=1,3
IN = ELNO(IP,IE)
IO = NF*IP - NF
II0 = NF*IN - NF
DO 100 JP=1,3

JN = ELNO(JP,IE)
JO = NF*JP - NF
JJO = NF*JN - NF

C MERGE PARTITION INTO GLOBAL MATRIX
DO 50 I=1,NF
DO 50 J=1,NF
50 KMAT(II0+I,JJO+J) = KMAT(II0+I,JJO+J) + K(I0+I,J0+J)
100 CONTINUE

C ADD CONSTRAINED ROWS AND COLUMNS TO INDEPENDENT ROWS AND COLUMNS
DC 200 ILOC=1,NN
IILOC = KFD(ILOC)
IF(IILOC.LT.0)IILOC = -IILOC
IF(IILOC.EQ.0)GO TO 200
DO 180 I=1,NN
180 KMAT(I,IILOC) = KMAT(I,IILOC) + KMAT(I,ILOC)
DO 190 J=1,NN
190 KMAT(IILOC,J) = KMAT(IILOC,J) + KMAT(ILOC,J)
200 CONTINUE
RETURN
END

SUBROUTINE DECOMP(KMAT,NN,KFD,IDE,DET)

DECOMPOSE GLOBAL STIFFNESS MATRIX KMAT, AND COMPUTE DETERMINANT.

KMAT = GLOBAL (TOTAL SYSTEM) MATRIX, WHICH MAY BE UNSYMMETRIC.
NN = TOTAL SYSTEM DOF.
KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.
KFD(I) = +,- = SPECIFIED FORCE,DISPLACEMENT.
KFD(I) = +-J INDICATES INDEPENDENT DOF.
KFD(I) = +-J INDICATES DEPENDENT FREEDOM WHOSE DISPLACEMENT IS CONSTRAINED TO EQUAL DISPLACEMENT AT J.
IDENT,DET = SIGN,LOGARITHM (BASE 10) OF DETERMINANT.

INTEGER NN,KFD(1),IDENT
DOUBLE PRECISION KMAT(NN,NN),DET
INTEGER NM1,JJ,J1,J2,II,J
DOUBLE PRECISION C,C
IF(NN.EQ.1)RETURN
NM1 = NN-1
DO 200 II=1,NM1
IF(KFD(II).NE.II)GO TO 200
DO = 1.00/KMAT(II,II)  
JJ1 = II+1  

C DISTRIBUTE II TH ROW OF UPPER TRIANGULAR MATRIX  
CC 90 J=JJ1,NN  
C = O*KMAT(II,J)  
IF(C.EQ.0.DO)GO TO 90  

C DISTRIBUTE KMAT(II,J) TO JTH COLUMN  
DC 50 I=JJ1,J  
50 KMAT(I,J) = KMAT(I,J) - C*KMAT(I,II)  
90 CONTINUE  

C DISTRIBUTE II TH COLUMN OF LOWER TRIANGULAR MATRIX  
JJ2 = II+2  
IF(JJ2.GT.NN)GO TO 200  
DC 190 J=JJ2,NN  
C = D*KMAT(J,II)  
IF(C.EQ.0.DO)GO TO 190  

C DISTRIBUTE KMAT(J,II) TO JTH ROW  
JM1 = J-1  
150 DO 150 I=JM1,JMI  
150 KMAT(J,I) = KMAT(Jl) - C*KMAT(I,II)  
190 CONTINUE  
200 CONTINUE  

C COMPUTE DETERMINANT  
ICET = 1  
DET = 0.00  
DO 500 II=1,NN  
IF(KFD(II).NE.I)GO TO 500  
C = KMAT(II,II)  
IF,ID. LT.0.DO)IDET = -IDET  
DET = DET + DLOG10(DABS(D))  
500 CONTINUE  

1201 FORMAT(1HO,'JACOBIAN DETERMINANT SIGN,LOGARITHM =',15,D15.7)  
WRITE(6,1201)IDET,DET  
RETURN  
END

SUBROUTINE SOLVE(KMAT,NN,KFD,P,Q)  
C FCRWARC-BACK SUBSTITUTE TO SOLVE FOR FORCES P AND DISPLACEMENTS Q.  
C KMAT = DECOMPOSED SYSTEM JACOBIAN MATRIX (MAY BE UNSYMMETRIC).
NN = TOTAL SYSTEM DOF.

KFD(I) = FORCE-DISPLACEMENT-CONSTRAINT SPECIFICATION FOR DOF I.
KFD(I) = +,- = SPECIFIED FORCE,DISPLACEMENT.
KFD(I) = +/- INDICATES INDEPENDENT DOF.
KFD(I) = +/-J INDICATES DEPENDENT FREEDOM WHOSE DISPLACEMENT IS
      CONSTRAINED TO EQUAL DISPLACEMENT AT J.
P(I),Q(I) = FORCE,DISPLACEMENT AT COF I.

EACH DOF HAS EITHER SPECIFIED FORCE OR DISPLACEMENT.
INPUT P = SPECIFIED FORCES AND DISPLACEMENTS, Q = GARBAGE.
OUTPUT P = SPECIFIED AND COMPUTED FORCES, Q = SPECIFIED AND
      COMPUTED DISPLACEMENTS.

INTEGER NN,KFD(I)
DOUBLE PRECISION KMAT(NN,NN),P(I),Q(I)
INTEGER I,J,II,IDUM
DOUBLE PRECISION C

CORRELATE DEPENDENT-INDEPENDENT DOFS, ZERO OUT Q
DO 5 I=1,NN
   J = IABS(KFD(I))
   IF(J.EQ.I)GO TO 5
   IF(KFD(J).GT.0)P(J) = P(J) + P(I)
   P(I) = 0.00
   Q(I) = 0.00
5 Q(I) = 0.00

FORWARD SUBSTITUTION
DO 100 I=1,NN
   IF(KFD(I).NE.1)GO TO 31
   C = (Q(I)+P(I))/KMAT(I,I)
   Q(I) = C
   GC TO 41
31 C = P(I)
   P(I) = -Q(I)
   Q(I) = C
41 IF(I.EQ.NN)GO TO 100
   II = I+1
   GC 50 J=II,NN
50 Q(J) = Q(J) - KMAT(J,I)*C
100 CONTINUE

BACKWARD SUBSTITUTION
I = NN+1
DO 200 IDUM=1,NN
   I = I-1
C = 0.0D0
IF(KFD(I).NE.I)GO TO 131
IF(I.EQ.NN)GO TO 200
I1 = I1 + 1
DC 120 J = I1, NN
120 C = C + KMAT(I,J) * Q(J)
Q(I) = Q(I) - C / KMAT(I, I)
GC TO 200
131 DC 140 J = I1, NN
140 C = C + KMAT(I, J) * Q(J)
P(I) = P(I) + C
200 CONTINUE
C
SET DEPENDENT DISPLACEMENTS, AND ADJUST INDEPENDENT FORCES
DC 205 I = I1, NN
J = IABS(KFD(I))
IF(J.EQ.I)GO TO 205
Q(I) = Q(J)
P(J) = P(J) - P(I)
205 CONTINUE
RETURN
END

START 5 5 6
TORUS MESH-12 (R=10, R=2, T=0.05) MCONEY MATERIAL, C1=80, C2=20 K. VOS
3 2 1 5 1.0-6
1 1.C-3 1.0-6
2 .01 1.0 0.1 -10.0
1 .01 3.05
2 80.0 20.0

1 1 8.0 0.0 0.0 1
2 1 8.0 2.0 0.0 1
3 1 8.06814830.0 .517631257 1
4 1 8.06814832.0 .517631257 1
5 1 8.26794920.0 1.0 1
6 1 8.26794922.0 1.0 1
7 1 8.58578640.0 1.41421356 1
8 1 8.58578642.0 1.41421356 1
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