THE NASA HIGH-POWER CARBON DIOXIDE LASER - A VERSATILE TOOL FOR LASER APPLICATIONS

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Abstract

The NASA Lewis Research Center has designed and fabricated a closed-cycle, continuous wave (CW) carbon dioxide (CO₂) high-power laser to support research for the identification and evaluation of possible high-power laser applications. The device is designed to generate up to 70 kW of laser power in annular-shape beams from 1 to 9 cm in diameter. Electric discharge, either self-sustained or electron-beam-sustained, is used for excitation. This laser facility can be used in two ways. First, it provides a versatile tool on which research can be performed to advance the state-of-the-art technology of high-power CO₂ lasers in such areas as electric excitation, laser chemistry, and quality of output beams, all of which are important whether the laser application is government or industry oriented. Second, the facility provides a well-defined, continuous wave beam for various application experiments, such as propulsion, power conversion, and materials processing.

Introduction

The National Aeronautics and Space Administration (NASA) is engaged in a high-power laser research program. The general objective of this program is to define and evaluate the potential of high-power lasers for NASA applications. The Lewis Research Center of NASA is responsible for the systems definition portion of this program. In support of this broad technological responsibility, research and development activities (contractual and in-house) are conducted in key areas such as advanced lasing concepts, transmission of laser energy over long distances and its ultimate conversion into other forms, and the advancement of CO₂ laser technology in areas of interest to NASA. All of these areas have resulted from possible NASA mission scenarios which dictated the following basic laser system requirements:

(1) Multimegawatt laser devices
(2) CW operation for duration of several minutes to indefinite periods
(3) Completely closed-cycle operation (to eliminate storables in space)
(4) Large adaptive optics systems for transmission of high power over thousands of kilometers

The second and third requirements are not unique to NASA. Industrial users of multikilowatt lasers, faced with efficiency and economy considerations, will eventually require completely closed-cycle devices. Although NASA-Lewis has a significant interest in shorter wavelength lasers, carbon dioxide lasers offer an existing level of technological maturity which will serve as a reference for future high-power lasers operating at different wavelengths. It is for this reason that NASA-Lewis has designed and fabricated a CO₂, electric-discharge, closed-cycle, laser system which generates a beam with a wavelength of 10.6 μm. This system not only supplies a high-power beam for application experiments, but also provides the capability for investigating some basic parameters of CO₂ lasers which previously have been cursorily treated. This paper covers the general design of the high-power laser facility, some preliminary data, and operating characteristics.

Design Criteria

NASA's interest in CO₂ lasers is based on the fact that carbon dioxide - nitrogen - helium (CO₂-N₂-He) is the only high-power lasing medium which has been developed to the extent that it could satisfy the criteria just listed. Even then, there are characteristics of this medium which are not clearly defined, e.g., the feasibility for use on future NASA missions may be determined. The following areas are of particular interest:

(1) Optimum electric excitation method for long-duration, high-power operation
(2) System efficiency
(3) Chemical polconlni effects on gas mixtures
(4) Optimum optical cavity configuration effects for best beam quality

The goal of the design of the NASA laser is to have a facility with the capability for investigating these areas and simultaneously conducting beam application experiments.

Excitation Method

Electric excitation has been considered the best method of exciting the CO₂ laser primarily because the efficiency of an electric discharge laser is apparently higher than that of a similar gas dynamic laser. However, the optimum method of electric excitation for long-duration continuous-high-output power remains to be determined. In any method, the major problem is one of controlling the level of electron generation. Electron-beam-sustained discharges have been promoted(1,2) as the best method of independent control of electron generation and electron energy in the discharge. This method requires large highly reliable electron transmission foils for long-duration operation together with very-high-voltage power supplies. Radiofrequency-stabilized, self-sustained discharges have been promoted(3) to be reasonably successful, and also ultraviolet, preionized, self-sustained discharges have been suggested(4) as a possible solution to the electron avalanche control problem. An innovative solution to this problem has been suggested(5) in which a properly tailored and operated 50-electron-volt discharge controls the electron avalanche process and maintains a steady-state plasma condition. This method, called impulse ionization, requires highly sophisticated pulse generating circuits. The most straightforward approach to electric excitation is a pure, self-sustained discharge. A question exists, however, of just how far this type of discharge can be scaled before the onset of instabilities.
The NASA high-power laser has been designed with a cavity which allows more than one type of discharge to be investigated in the same device. The pin–plane, self-sustained discharge (Fig. 1(a), was chosen as the type to be evaluated first, and the electron-beam-sustained discharge (Fig. 1(b)) will be evaluated second. Output laser powers up to 70 kW are expected from the facility.

Gas Contamination

One of the major factors which have limited the operation of closed-cycle, CW, high-power lasers has been the effect of gas contamination due to plasma-induced chemical reactions in the lasing gases. To circumvent this problem, industrial type, multikilowatt CO₂ systems (1, 3) supply makeup gases of up to 10 percent of the mixture of CO₂, N₂, and He to cleanse the gases. This makeup gas can substantially increase the cost of operating a system over an extended period, and furthermore require added gas storage space. Figure 2 shows some of the various reaction products that can occur in an electrically excited lasing mixture of CO₂, N₂ and He. It was shown in the work of reference 7 that the nitrogen-oxygen compounds were the worst contaminants. Results of that controlled experiment on a small, low-pressure (650 to 2000 N/m² (5 to 15 torr)) laser amplifier in which various contaminants were added are shown in Table 1. They indicate only 400 ppm of NO₂ can cause plasma instabilities, with complete loss of gain occurring with 1000 ppm. The NASA laser has been designed to assess and investigate contamination effects by using low-outgassing materials and high-vacuum technology. It is instrumented to monitor gas mixture compositional changes during operation. One of the goals is to determine the means of neutralizing the effects of contaminants.

Optical Configuration

Laser beam propagation is directly affected by the quality of the laser device output beam. The NASA laser has been designed to accommodate optics of various sizes in order to investigate the combined effects of beam size, optical paths, and electric discharge type on the output beam quality, both spatial and temporal. Optical configurations using single- and multiple-pass unstable oscillators or power amplifiers or both are possible in the device.

Description of Laser

A drawing of the NASA high-power laser is shown in Figure 3. The gas flow loop occupies a flow space of approximately 6 by 6 m (20 by 20 ft) and contains a flow volume of approximately 20 m³ (700 ft³). The major components are (1) the vacuum system and gas supply system (not shown in the figure), (2) the pressure blower, (3) the laser test cavity, (4) the diffuser, and (5) the heat exchanger. Each of these components is described in this section. The laser system operating specifications are listed in Table 2.

Vacuum and Gas Fill System

The vacuum system is used to evacuate the flow loop to below atmospheric pressure before the system is filled with the lasing gas mixture of CO₂, N₂, and He. A unique feature of this facility is the capability of obtaining a hard vacuum. The low background pressure, together with the use of stainless steel and other low-outgassing materials in the loop, removes from consideration the effect of residual gases causing contamination of the lasing gases. Three stages of vacuum pumping, two roughing systems, and a 50-cm (20-in) oil-diffusion pump allow the laser loop to be evacuated to a level of 1.3x10⁻⁶ N/m² (2x10⁻⁵ torr). The roughing systems alone can maintain the loop at less than 6.13 N/m² (1.0 μm Hg), has.

The gas fill system, located outside the building, consists of three large-capacity, high-pressure storage tanks for industrial grade CO₂, N₂, and He. The gases are introduced into the laser flow loop through the plenum tank.

Pressure Blower

The centrifugal blower has an impeller 1.83 m (6 ft) in diameter. Although designed and fabricated to NASA specifications, the general design is similar to common industrial exhaust blowers. An unusual characteristic, however, is the requirement that it operate over a pressure range from 1.3x10³ to 1.3x10⁹ N/m² (100 to 800 torr) and circulate gases at various densities and velocities around the laser loop. The velocity design point was 100 m/sec. The fractional amount of helium in the gas mixture, which varies from 50 to 80 percent, places the most restriction on the blower performance. The blower is driven through a variable-speed fluid coupling by a 250-hp motor.

Heat Exchanger and Diffuser

The NASA laser loop utilizes only one heat exchanger as opposed to the two that are found in most closed-cycle systems. The principle behind this design is that the gas is overcooled prior to entering the blower, which then does work on the gas to raise its temperature up to the control point, nominally room temperature. The NASA heat exchanger has two stages within one housing. The first stage uses water from a package cooling tower, and the second stage uses cold gas. The gas is admitted to the second stage as a two-phase mixture.
second stage uses an ethylene glycol-water solution from a package chiller system. The heat exchanger coils are standard industrial air-conditioning coils consisting of swaged aluminum fins on copper tubes. They have a frontal area of approximately 5 m² (52 ft²).

Total heat removal capacity is 650 kW while a 20°C gas temperature is maintained in the plasma flow loop. The coils are being outgassed by passing a 90°C fluid through them.

Since the blowers system must operate over a large range of pressures and densities, there are some conditions at which the pressure head developed is limited. Therefore, a diffuser, installed downstream of the laser test cavity, was designed to recover as much as 80% of the pressure lost in the test cavity. The diffuser inlet has a cross section which is adjustable to match that of the test cavity.

Laser Test Cavity

The laser test cavity is shown in Figure 4. Flexibility is the key word in describing this cavity, in which the directions of the discharge, optics, and flow are mutually orthogonal. The cavity consists of four major components: the main frame and bellmouth section, the cathode section, the anode section, and the optics.

The main frame is a structure which connects the electrode sections, mirror mounts, and flow loop. It is approximately 1.4 m (4.5 ft) in the flow direction by 1.5 m (5.0 ft) in the optics direction. The open-side design allows different electrode configurations and spacings to be inserted in the top and bottom openings, as well as a variety of optical configurations. Spacing is controlled by inserting spacers resembling picture frames between the cathode and anode sections and the main frame. These spacers allow interelectrode spacings (and the optical cross section) to be adjusted from 1 to 9 cm in 2-cm increments. Inserted into the main cavity frame are two-dimensional bellmouth plates which can also be adjusted to the cathode-anode spacing. Their purpose is to aerodynamically tailor the gas stream entering the test cavity. The main frame is electrically isolated from the rest of the flow loop by fiberglass-epoxy spacers. The internal side walls are also lined with fiberglass and epoxy to minimize arc attachment to bare metal.

There are five optics ports on each side of the main frame. These ports are 12 cm in diameter, and their centerlines are separated by 16.5 cm. Thus, single- or multiple-pass optical configurations are possible.

The cathode section consists of a stainless-steel frame 1 by 1.5 m, which is peculiar to the type of electric discharge being used. That is, the pin-to-plane self-sustained discharge assembly has its own cathode frame, as does the electron-beam-sustained discharge assembly. The pin-to-plane cathode frame is grid-shaped, which allows for five independent channels for electric excitation. The surfaces of this frame, exposed to the discharge, are coated with 0.5-mm (20 mils) of epoxy applied by using a fluidized-bed technique. Ten fiberglass-epoxy panels fit into these grid openings, two per channel. The electrodes (pins) are epoxied into the panels in rows and columns forming a hexagonal array with one pin per cm². The pins are 1.5 mm in diameter and are made of tungsten - 3 percent rhenium. The tips, which are pointed, protrude into the gas flow 1 cm for cooling purposes. This means the flow gap is 1 cm larger than the interelectrode spacing.

In any one channel, there are 21 rows of pins with 74 pins per odd-number row and 72 pins per even-number row. Currently, each pin is ballasted with a 20-KΩ resistor, and each row of pins is ballasted from zero to 4500 0. The row ballast can be varied remotely to tailor the discharge current in the flow direction.

The cathode frame assembly for the electron-beam-sustained discharge allows for one channel of electric excitation. This assembly, shown in Figure 5, consists of a cathode plate and an electron gun. The cathode plate is a support structure for the screen cathode and the electron foil window. This window, which is typically 0.025 cm (1.0 mil) thick aluminum, serves as an electron-transparent divider between the laser gases and the high vacuum in the electron gun. The electron gun itself is in a large vacuum tank, which houses the hot-filament electron sources, control grid, and screen anode. It is supported by the cathode plate. The filament and control grid assembly are electrically isolated from the vacuum tank and float at a high negative voltage. The electron gun will typically provide continuously 0.15 mA/cm² of 175-kV electrons through the 10- by 125-cm foil window. The electron-beam device is used to provide the electrons and ionization necessary to run a discharge between the laser-cavity anode and cathode by means of a separately applied electric field.

The anode frame section is essentially the same design for both methods of electric excitation. It consists of a fiberglass-epoxy plate, identical in size to the cathode frame, and has embedded in it an oil-cooled copper anode. The anode used for the pin-to-plane discharge is large enough to cover the three middle channels of the cathode frame and is 56 by 135 cm. The copper anode for the electron-beam-sustained discharge is 20 by 135 cm.

The basic optical configuration used in the laser cavity forms a single-pass unstable resonator. Possible variations of this basic configuration include folded unstable resonators and oscillator-amplifier combination. Magnifications from 1.25 to 1.81 are currently available for use on single-pass configurations with a 2.1-cm optical path length. Depending on the particular interelectrode spacing and separator mirror hole size in use, these optics provide equivalent Fresnel numbers of 0.5 to 9.5 and output beam diameters from 2.4 to 6.3 cm. The mirrors, which are water-cooled copper, are also remotely adjustable to maintain the beam direction. The mirrors can be remotely adjusted on two axes by using stepping motors. The beam exits vertically through a zinc selenide window and is redirected to a horizontal position by an additional water-cooled flat copper mirror, which is also remotely adjustable.
Instrumentation and Control

The vacuum, gas fill, blower, and heat exchanger subsystems are controlled and sequenced during operations by a programmable controller. This device not only serves a labor-saving function, but also aids in trouble shooting and facilitates the changing of modes of operation. High-voltage controls and sequencing have been hard-wired for reasons of safety. Bulk laser controls and measurements are performed by using a computerized electronic monitoring system which employs a high-speed data-acquisition system.

Bulk laser control and measurement systems are performed by using a computerized electronic monitoring system which employs a high-speed data-acquisition system. The system is capable of monitoring and controlling the entire laser system, including gas injection, pressure control, and optical alignment. The system is also capable of performing electronic monitoring of the laser system, including gas injection, pressure control, and optical alignment.
Concluding Remarks

A description of a versatile, continuous wave, completely closed-cycle CO₂ laser has been presented. The initial operation of the device has indicated that a self-sustained, pin-to-plane discharge can generate many kilowatts of laser power. After complete evaluation of this type of discharge plus the implementation of an electron-beam-sustained discharge, the laser should provide means of generating power in the 50- to 70-kW range. The simplistic nature of the pin-to-plane design offers, with sufficient development, advantages over the electron-beam system, particularly for output power in the 1- to 10-kW range. The major advantage is that the requirement for an extended life, maintenance free, electron gun system is eliminated. However, the pin-to-plane system does require considerable fabrication and installation labor.

The flexibility of the NASA high-power laser not only allows the capability of investigation various excitation techniques, gas contamination effects, and optical configurations, but also permits experiments in support of the NASA research programs in the areas of adaptive optics, laser propulsion, laser power conversion, and materials processing.

References

Table 1. Nitrogen-Oxygen Contamination Effects(7)

<table>
<thead>
<tr>
<th>Contaminant</th>
<th>Effect</th>
</tr>
</thead>
<tbody>
<tr>
<td>NO</td>
<td>Gain and plasma impedance began to decrease at 0.1% (1000 ppm); complete loss of gain at 1.5% (15 000 ppm)</td>
</tr>
<tr>
<td>NO&lt;sub&gt;2&lt;/sub&gt;</td>
<td>0.04% (400 ppm) caused plasma instabilities; complete loss of gain at 0.1% (1000 ppm)</td>
</tr>
<tr>
<td>N&lt;sub&gt;2&lt;/sub&gt;O</td>
<td>0.08% (800 ppm) caused plasma instabilities; complete loss of gain at 0.2% (2000 ppm)</td>
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</table>

Table 2. NASA High-Power-Laser Specifications

(a) Flow loop

<table>
<thead>
<tr>
<th>Parameter</th>
<th>Specification</th>
</tr>
</thead>
<tbody>
<tr>
<td>Pressure range, N/m&lt;sup&gt;2&lt;/sup&gt; (torr)</td>
<td>1.33x10&lt;sup&gt;-1&lt;/sup&gt;-1.1x10&lt;sup&gt;5&lt;/sup&gt; (100-800)</td>
</tr>
<tr>
<td>Vacuum capability, N/m&lt;sup&gt;2&lt;/sup&gt; (torr)</td>
<td>1.2x10&lt;sup&gt;-4&lt;/sup&gt; (10&lt;sup&gt;-5&lt;/sup&gt;)</td>
</tr>
<tr>
<td>Velocity range, m/sec</td>
<td>50-150</td>
</tr>
<tr>
<td>Mass flow range, kg/sec</td>
<td>1.0-12.5</td>
</tr>
<tr>
<td>Gas mixture, CO&lt;sub&gt;2&lt;/sub&gt;, N&lt;sub&gt;2&lt;/sub&gt;, He</td>
<td>50%&lt;He&lt;80%</td>
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<tr>
<td>Heat exchanger capacity (two stages), kW</td>
<td>650</td>
</tr>
<tr>
<td>Cavity flow area, cm&lt;sup&gt;2&lt;/sup&gt;</td>
<td>2 to 10 by 150</td>
</tr>
</tbody>
</table>

(b) Test cavity

<table>
<thead>
<tr>
<th>Pin-to-plane discharge</th>
<th>Electron-beam discharge</th>
</tr>
</thead>
<tbody>
<tr>
<td>Channels for excitation and optics</td>
<td>Channels for excitation and optics</td>
</tr>
<tr>
<td>Channels currently in use</td>
<td>5</td>
</tr>
<tr>
<td>Pin-to-plane discharge</td>
<td>1</td>
</tr>
<tr>
<td>Cathode pin diameter, mm</td>
<td>5.5</td>
</tr>
<tr>
<td>Pin spacing, Pins/cm&lt;sup&gt;2&lt;/sup&gt;</td>
<td>1</td>
</tr>
<tr>
<td>Pin spacing, Pins/row</td>
<td>72 or 74</td>
</tr>
<tr>
<td>Pin spacing, Pins/row/channel</td>
<td>21</td>
</tr>
<tr>
<td>Cathode ballast</td>
<td>20</td>
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<tr>
<td>Pin resistors, kΩ</td>
<td>0 to 1500</td>
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<tr>
<td>Row resistors, kΩ</td>
<td>20</td>
</tr>
<tr>
<td>Size of anode (oil-cooled copper plate), cm</td>
<td>56 by 135</td>
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<tr>
<td>Interelectrode spacing, cm</td>
<td>1 by 9</td>
</tr>
<tr>
<td>Maximum current, A</td>
<td>30</td>
</tr>
<tr>
<td>Maximum applied voltage, kV</td>
<td>20</td>
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<tr>
<td>Maximum excited area per channel, cm&lt;sup&gt;2&lt;/sup&gt;</td>
<td>0.12 by 125</td>
</tr>
</tbody>
</table>
Figure 1. Laser discharge schematics.

(a) Pin-to-plane self-sustained.  (b) Electron beam sustained.

Figure 2. Plasma-induced chemical reactions.
Figure 3. - NASA high-power laser.
Figure 4. - Laser test cavity with pin-to-plane self-sustained discharge.
Figure 5. - Laser test cavity with electron-beam-sustained discharge.
Figure 6. - Discharge power density and distribution of ratio of electric field to neutral density. Laser power, 5.1 kW; discharge power, 74.0 kW; ballast power, 24.3 kW; total current, approximately 14.78 A; gas velocity, 104.9 m/sec; pressure, 147.3 torr.

Figure 7. - Laser performance.