Tensile Stress-Strain Behavior of Boron/Aluminum Laminates

J. A. Sova and C. C. Poe, Jr.

FEBRUARY 1978
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J. A. Sova
The George Washington University
Joint Institute for Advancement of Flight Sciences
Langley Research Center
Hampton, Virginia

and

C. C. Poe, Jr.
Langley Research Center
Hampton, Virginia

NASA
National Aeronautics
and Space Administration

Scientific and Technical
Information Office

1978
SUMMARY

The tensile stress-strain behavior of five types of boron/aluminum laminates was investigated. Two of the laminate types contained only 0° or 45° fibers ([0]_T and [+45]_S); the remaining three consisted of both 0° and 45° plies ([-45], [0/45], and [+45/0]). Longitudinal and transverse stress-strain curves were obtained for monotonic loading to failure and for three cycles of loading to successively higher load levels.

For the laminates with 0° plies in the loading direction, the ultimate strengths varied linearly with the percentage of 0° plies in the composite. The strengths predicted by assuming that the 0° plies failed first correlated well with the experimental results. The stress-strain curves for all the boron/aluminum laminates were nonlinear except at very small strains. Within the small linear regions, elastic constants calculated from laminate theory corresponded to those obtained experimentally to within 10 to 20 percent. However, because of the nonlinearity, the elastic constants do not characterize the laminates over a significant part of their working range. The nonlinearity in the stress-strain curves was greater at higher percentages of 45° plies in the laminate.

A limited amount of cyclic loading did not affect the ultimate strength and strain for the boron/aluminum laminates. The laminates, however, exhibited a permanent strain on unloading. As with the nonlinearity in the stress-strain curves, the amount of permanent strain was greater for laminates with more 45° plies.

The Ramberg-Osgood equation was fitted to the stress-strain curves to obtain average curves for the various laminates. The equation fitted the experimental data well.

INTRODUCTION

The analysis of structures made from boron/aluminum laminates is complicated by the inelasticity and nonlinearity of the stress-strain behavior in the working range of the material. Although the mechanical properties of boron/aluminum laminates have been studied (see, for example, refs. 1, 2, and 3), the stress-strain behavior has not been adequately characterized.

The objective of this investigation was to characterize the stress-strain behavior of several types of boron/aluminum laminates. Two of the laminate types consisted of only 0° or 45° plies; three other laminate types contained both 0° and 45° plies. Both longitudinal and transverse stress-strain curves were obtained for the various laminates. To investigate the inelasticity of the material, some specimens were also subjected to a few loading cycles. The
Ramberg-Osgood equation was fitted to the experimental stress-strain data. Elastic constants and the extent of the linear regions in the stress-strain curves were determined for the various laminates. Elastic constants calculated from laminate theory were compared with measured values.

SYMBOLS

Although values are given in the International System of Units (SI) in this report, the experimental procedures were carried out using U.S. Customary Units.

- \( b \) constant in the Ramberg-Osgood equation, Pa\(^{-1}\)
- \( E \) Young's modulus of elasticity, Pa
- \( E_f \) Young's modulus of elasticity of fibers, Pa
- \( E_s \) secondary modulus, Pa
- \( E_{\text{tan}} \) tangent modulus, Pa
- \( F_{\text{tu}} \) ultimate tensile strength, Pa
- \( G_{12} \) lamina shear modulus, Pa
- \( G_{xy} \) laminate shear modulus, Pa
- \( g \) acceleration due to gravity, m/sec\(^2\)
- \( n \) constant in the Ramberg-Osgood equation
- \( V_f \) fiber volume fraction
- \( \Gamma \) percentage of 45° plies
- \( \gamma \) shear strain
- \( \epsilon \) axial strain
- \( \epsilon_{\text{tu}} \) ultimate tensile strain
- \( \Lambda \) percentage of plies with fibers in loading direction
- \( \nu \) Poisson's ratio
- \( \nu_{\text{tan}} \) tangent Poisson's ratio
- \( \sigma \) axial stress, Pa
- \( \sigma_\epsilon \) axial stress at a strain of \( \epsilon \), Pa
\( \tau \) shear stress, Pa

Subscripts:

\( x, y \) Cartesian coordinates

\( \alpha, \beta, \eta, \xi \) refer to legs of strain gage rosettes (see fig. 2)

The notation for laminate orientation used in reference 4 is used in the present report. The cross-ply angles are listed in the order of layup, separated by a slash, with the entire listing enclosed within brackets. Where there is more than one consecutive lamina at any given angle, the number of laminae at that angle is denoted by a numerical subscript within the brackets. Subscript S outside the brackets denotes symmetric; for example, \([0/\pm45]_S\) means \([0/45/-45/-45/+45/0]\). Subscript T outside the brackets denotes total; for example, \([0]_{6T}\) means \([0/0/0/0/0/0]\).

**EXPERIMENTAL PROCEDURES**

**Material and Specimens**

A diffusion-bonded boron/aluminum sheet material consisting of 0.142-mm-diameter boron fibers embedded in 6061 aluminum matrix was used in this investigation. The material, which was fabricated for a larger fracture program, was tested in the "as fabricated" condition. Ultrasonic (C-scan) inspection did not reveal any voids in the material. As shown in the following table, 60 sheets of the material, which included 5 types of laminates were used. The sheets were delivered in two sizes: 508 by 457 mm and 508 by 305 mm. The 0° fiber direction was parallel to the 508-mm sides. (See fig. 1.) The fiber volume fraction, specified to be 45 to 50 percent, was higher for the unidirectional laminates than for the laminates with cross plies, as indicated in the table. The average sheet thickness listed in the table was calculated by averaging measured thicknesses of all the sheets of a single laminate orientation.

<table>
<thead>
<tr>
<th>Sheet number</th>
<th>Laminate orientation</th>
<th>Average sheet thickness, mm</th>
<th>Number of plies</th>
<th>Average ply thickness, mm</th>
<th>Volume fraction, percent</th>
</tr>
</thead>
<tbody>
<tr>
<td>1 to 10</td>
<td>([0]_{6T})</td>
<td>1.068</td>
<td>6</td>
<td>0.178</td>
<td>50</td>
</tr>
<tr>
<td>11 to 25</td>
<td>([\pm45]_{2S})</td>
<td>1.528</td>
<td>8</td>
<td>0.191</td>
<td>45</td>
</tr>
<tr>
<td>26 to 40</td>
<td>([0/\pm45]_S)</td>
<td>1.110</td>
<td>6</td>
<td>0.185</td>
<td>45</td>
</tr>
<tr>
<td>41 to 55</td>
<td>([0_2/\pm45]_S)</td>
<td>1.488</td>
<td>8</td>
<td>0.186</td>
<td>44</td>
</tr>
<tr>
<td>56 to 60</td>
<td>([\pm45/0_2]_S)</td>
<td>1.512</td>
<td>8</td>
<td>0.189</td>
<td>44</td>
</tr>
</tbody>
</table>
One longitudinal and one transverse tensile specimen were tested from each sheet of material. (See fig. 1.) The specimens were of simple rectangular shape, 19.0 mm wide and 254.0 mm long, and were numbered according to the following four-digit system: the first two digits denote the sheet number, the third digit denotes the position of the specimen within the original sheet, and the fourth digit indicates whether the specimen was longitudinal or transverse.

Test Procedure and Equipment

The specimens were tested in a hydraulically actuated, closed-loop, servo-controlled testing machine with hydraulic grips. Plastic tabs about 3 mm thick were placed between the grips and the specimens to protect the surface of the specimens. Radiographs showed that the gripping pressure necessary to hold the specimens did not damage the boron fibers. No specimens failed in the grips.

Loads for the static tensile tests were programmed by an analog ramp generator to vary linearly with time. The specimens were loaded to failure slowly (in about 2 minutes) so that at least 50 load-strain readings could be recorded during each test.

For the cyclic tensile tests, the specimens were loaded to 0.267 of the static ultimate strength, unloaded, reloaded to 0.467 of the ultimate strength, unloaded, reloaded to 0.667 of the ultimate strength, unloaded, and finally loaded to failure. The values of 0.267 and 0.667 of the ultimate strength represent the upper values of a 1g load (transport-type airplane) and of a limit load, respectively; 0.467 of the ultimate strength is the average of the other two values.

Strain gage rosettes were used to measure strains in the direction of load and at angles of ±45° to the load direction. The gages were bonded to the center of the specimens on both front and back sides. The transverse and shear strains for each rosette were calculated from the measured strains as follows:

\[
\begin{align*}
\varepsilon_\eta &= \varepsilon_\alpha + \varepsilon_\beta - \varepsilon_\zeta \\
\gamma &= \varepsilon_\alpha - \varepsilon_\beta
\end{align*}
\]  

(1)

where the subscripts refer to the legs of the rosette shown in figure 2. The axial and shear strains on both sides of each specimen were averaged to eliminate any strains due to bending in the specimens. The shear strains were very small compared with the axial strains, indicating that the loads were applied without twist or shear.

During both the static and cyclic tests, the output signals from the strain gages, the load cell, and a peak-load reader were recorded on the digital data acquisition system shown schematically in figure 3. As shown in figure 2, the output signals were recorded on 12 channels, and the load was recorded 4 times during each scan. These analog output signals were scanned continuously at about 2 seconds per scan, were converted into digital signals, and then were printed on paper tape and recorded on magnetic tape. The scan number, the specimen number, and the time to the nearest second were also
recorded. Subsequently, the data on the magnetic tapes were analyzed by a
digital computer. The maximum load used to calculate the ultimate tensile
strength was determined from the peak-load reader.

Because the load varied while the strains were recorded, the computer pro-
gram used the following procedure to derive the stress-strain curves. The load
corresponding to a recorded strain was calculated by linearly interpolating
between loads recorded before and after the particular strain. The linear
interpolation was quite accurate because the load varied linearly with time and
the scanning rate was constant. The average sheet thickness for a particular
laminate orientation was used to calculate the corresponding stresses.

In calculating $E_y$ and $\gamma$ by equations (1), $\varepsilon_\alpha$ and $\varepsilon_\beta$ corresponding
to $\varepsilon_\gamma$ were calculated by linearly interpolating between values of $\varepsilon_\alpha$ and
$\varepsilon_\beta$ recorded before and after $\varepsilon_\gamma$. Because of the nonlinear response of the
material, the strains did not vary linearly with time as did the loads. Never-
theless, the linear interpolation was quite accurate because the change in
strain was very small between recordings.

STRESS-STRAIN RELATIONSHIP

Assuming the boron/aluminum laminates to be homogeneous materials with
orthotropic properties and in a state of plane stress, the linear strain-stress
relations are (ref. 5)

$$\varepsilon_y = \frac{\sigma_y}{E_y} - \frac{\sigma_x \gamma_{xy}}{E_x} \tag{2}$$

$$\varepsilon_x = \frac{\sigma_x}{E_x} - \frac{\sigma_y \gamma_{yx}}{E_y} \tag{3}$$

$$\gamma_{yx} = \frac{\tau_{yx}}{G_{yx}} \tag{4}$$

The laminate principal axes are referred to as $y$ and $x$ in this report, the
$y$-direction always being the direction of the 0° fibers. Four independent
elastic constants present in equations (2) to (4) characterize the laminates:
Young's moduli in the $y$- and $x$-directions, $E_y$ and $E_x$; the shear modulus $G_{yx}$;
and the major Poisson's ratio $\gamma_{yx}$. The fifth elastic constant, $\gamma_{xy}$, is
related to the other constants as follows:

$$\gamma_{xy} E_y = \gamma_{yx} E_x \tag{5}$$

The following form of the well-known Ramberg-Osgood equation (ref. 6) was
chosen to represent the tensile stress-strain curves of the boron/aluminum
material:
\[ \varepsilon = \frac{\sigma}{E} + (b\sigma^n) \]  \hspace{1cm} (6)

For orthotropic laminates, equation (6) can be written as follows for \( \varepsilon_y \) and \( \varepsilon_x \) for uniaxial loading in the y-direction,

\[
\begin{align*}
\varepsilon_y &= \frac{\sigma_y}{E_y} + (b_{yy}\sigma_y)^{n_{yy}} \\
\varepsilon_x &= -\frac{\sigma_y \nu_{yx}}{E_y} - (b_{yx}\sigma_y)^{n_{yx}}
\end{align*}
\]  \hspace{1cm} (7)

and for uniaxial loading in the x-direction,

\[
\begin{align*}
\varepsilon_y &= -\frac{\sigma_x \nu_{xy}}{E_x} - (b_{xy}\sigma_x)^{n_{xy}} \\
\varepsilon_x &= \frac{\sigma_x}{E_x} + (b_{xx}\sigma_x)^{n_{xx}}
\end{align*}
\]  \hspace{1cm} (8)

Equations (7) were fitted to stress-strain data for longitudinal specimens and equations (8) to data for transverse specimens. For linear-elastic behavior, the nonlinear term in equation (6) is zero, and equations (7) and (8) reduce to the uniaxial loading forms of equations (2) and (3). Note that because the constants in the nonlinear terms in equations (7) and (8) were derived from uniaxial tests, the sum of the equations may not give correct strains for biaxially applied stresses. The elastic constants \( E \) and \( \nu \) were calculated using only the linear-elastic part of the stress-strain curves, and the constants \( b \) and \( n \) were calculated using all of each curve. The procedure is given in the appendix.

RESULTS AND DISCUSSION

Static Tensile Tests

Stress-strain curves.- Stress-strain curves for the static tensile tests are presented in figures 4 to 8 for each laminate orientation. Each figure shows the measured strains in the y- and x-directions and average Ramberg-Osgood curves. The results for longitudinal and transverse specimens are shown in separate graphs in each figure. Note that for laminates containing 0° fibers, the stress scales for longitudinal and transverse tests in parts (a) and (b) of the figures differ by as much as a factor of 10. In general, the stress-strain curves are very nonlinear, and the Ramberg-Osgood curves model the experimental ones well. The curves are shown to failure except for the \([\pm 45]_{23} \) laminates where the strain gages debonded before failure. Tables I
to V present various parameters measured in the tests or calculated from the data, including the Ramberg-Osgood constants.

Figure 9 shows average Ramberg-Osgood stress-strain curves for all the laminates. The ultimate tensile strains of the four laminates with fibers in the direction of loading are about the same, 0.007 to 0.008. (See fig. 9(a).) In transverse tests of the four laminates, with some fibers normal to the loading direction, the ultimate tensile strains were much smaller and varied greatly, 0.0020 to 0.0065. For the \([45]_{2S}\) laminate orientations, the ultimate tensile strains were much higher than the 0.02 to 0.03 at which the gages failed; the ultimate strain given by the average Ramberg-Osgood equation for the average measured ultimate stress of 220.6 MPa is 0.05.

The stress-strain data show that the degree of nonlinearity is greater for higher proportions of 45° plies. In order to examine the nonlinearity in more detail, the tangent moduli and Poisson's ratios were calculated for both the test data and the average Ramberg-Osgood curves. The tangent moduli are given by

\[
\begin{align*}
(E_{\text{tan}})_x &= \frac{\partial \sigma_x}{\partial \epsilon_x} \\
(E_{\text{tan}})_y &= \frac{\partial \sigma_y}{\partial \epsilon_y}
\end{align*}
\]

and Poisson's ratios by

\[
\begin{align*}
(v_{\text{tan}})_{yx} &= -\frac{\partial \epsilon_x}{\partial \epsilon_y} \\
(v_{\text{tan}})_{xy} &= -\frac{\partial \epsilon_y}{\partial \epsilon_x}
\end{align*}
\]

The derivatives in equations (9) and (10) were calculated for the test data by taking the ratio of changes in stress and strain between successive readings.

Figures 10 to 14 show the applied stress plotted against the tangent modulus and Poisson's ratio for each laminate orientation. For laminates with some fibers in the loading direction, the curves for tangent modulus show that the laminates yielded at very low stresses, and the modulus approached a secondary value lower than the initial tangent modulus. If the laminate contained no fibers in the loading direction, the tangent modulus approached zero for high stresses. The low yield strength of the composite is due to the low yield strength of the aluminum matrix. With some fibers in the loading direction, the tangent moduli at high stresses agree well with those calculated by the following equation which assumes that the fibers in the loading direction carry all the load:
where \( \Lambda \) equals the percentage of plies with fibers in the loading direction. Values of \( V_f = 0.5 \) and \( E_f = 400 \text{ GPa} \) were used for the volume fraction and fiber modulus, respectively.

As previously noted, the overall conformity between the experimental stress-strain data and the approximating Ramberg-Osgood curves is good. However, figures 10 to 14 show that for laminates loaded in the direction of \( 0^\circ \) fibers, the shape of the tangent modulus curve given by the Ramberg-Osgood equation does not agree well with the data for large stresses, because the equation does not asymptotically approach a secondary modulus for large stress but rather approaches zero. Also, the shape of the curve does not agree well for small stresses near zero, because the derivative of the tangent modulus

\[
\frac{\partial E_{\text{tan}}}{\partial \sigma} = \frac{\partial}{\partial \varepsilon} \left( \frac{\partial \varepsilon}{\partial \sigma} \right)
\]

is equal to infinity rather than zero for \( \sigma = 0 \) and \( 1 < n < 2 \). For laminates not loaded in the direction of \( 0^\circ \) fibers, the shape of the tangent modulus curve given by the Ramberg-Osgood equation agrees well with the data, because the secondary modulus of the material is zero and \( n > 2 \).

For laminate orientations containing \( 0^\circ \) fibers, Poisson's ratio increases with stress for longitudinal tests and decreases with stress for transverse tests. For the \( [\pm 45]_2 \) laminate (fig. 14), Poisson's ratio increases dramatically with stress and approaches unity.

In order to determine the strains at the onset of the nonlinearity, the tangent moduli are plotted against strain in figure 15 for strains up to 0.0012. The plots show that the laminates yielded at a strain of about 0.00025 except for the \( [0]_{6T} \) laminates tested longitudinally. That laminate yielded at somewhat higher strains of about 0.0004, probably because the fibers are all in the loading direction and therefore cause virtually no concentration of strain in the matrix near the fibers. This yield strain is about equal to the yield strain of 0.0005 expected for the matrix itself.

Ultimate tensile strengths.- Figure 16 shows a comparison between experimental and predicted tensile strengths for the various boron/aluminum laminates. In the figure, ultimate tensile strengths are plotted against the percentages of \( 45^\circ \) plies in the laminates. The symbols are plotted at the average of the experimental results and the tick marks indicate the extremes. The difference between the strengths of \( [\pm 45/0]_S \) and \( [0_{-}/\pm 45]_S \) specimens indicates a small effect of stacking sequence on strength.

The theoretical lines (both solid and dashed) were derived by assuming that in both longitudinal and transverse tests, the \( 0^\circ \) plies fail first because
their failure strain is much smaller than that for the \(45^\circ\) plies. When failure of the \(0^\circ\) plies overloads the \(45^\circ\) plies, the laminate failure stresses are given by the solid lines in figure 16. For laminates with high percentages of \(45^\circ\) plies, however, the \(45^\circ\) plies are not overloaded and the laminates fail at strains and stresses higher than those at which the \(0^\circ\) plies fail; these failure stresses are shown by the dashed lines.

The equations for the lines were derived as follows. Assuming equal ply thicknesses, laminate theory gives the laminate stress in terms of the ply stresses as

\[
\sigma = (\sigma)_{0^\circ} - [(\sigma)_{0^\circ} - (\sigma)_{45^\circ}] \frac{\Gamma}{100}
\]

where \(\Gamma\) is the percentage of \(45^\circ\) plies and \((\sigma)_{0^\circ}\) and \((\sigma)_{45^\circ}\) are the stresses in the \(0^\circ\) and \(45^\circ\) plies, respectively. Thus, the laminate stress when the \(0^\circ\) plies fail is

\[
F_{tu} = (F_{tu})_{0^\circ} - [(F_{tu})_{0^\circ} - (\sigma E)_{45^\circ}] \frac{\Gamma}{100}
\]

where \((F_{tu})_{0^\circ}\) is the ultimate tensile strength of the \(0^\circ\) plies and \((\sigma E)_{45^\circ}\) is the stress in the \(45^\circ\) plies at the strain corresponding to failure of the \(0^\circ\) plies. Similarly, with the \(0^\circ\) plies having failed first, the laminate stress when the \(45^\circ\) plies fail is given by equation (12) with \((\sigma)_{0^\circ} = 0\) and \((\sigma)_{45^\circ} = (F_{tu})_{45^\circ}\), where \((F_{tu})_{45^\circ}\) is the ultimate tensile strength of the \(45^\circ\) plies. Thus,

\[
F_{tu} = (F_{tu})_{45^\circ} \frac{\Gamma}{100}
\]

The strength of the laminates, therefore, is given by the larger of the values calculated from equations (13) and (14). The values of \(F_{tu}\) and \(\sigma E\) given in the following table were used to calculate the lines in figure 16. The solid and dashed lines are given by equations (13) and (14), respectively. Average properties of the \([0]_{6T}\) and \([\pm45]_{2S}\) laminates in tables I and V were used for the \(0^\circ\) and \(45^\circ\) ply properties, respectively, and \(\sigma E\) for the \(45^\circ\) plies was determined from the average Ramberg-Osgood equations for the \([\pm45]_{2S}\) laminates. The agreement between the predicted and experimental results in figure 16 is good.

<table>
<thead>
<tr>
<th></th>
<th>((F_{tu})_{0^\circ}, \text{MPa})</th>
<th>((\epsilon_{tu})_{0^\circ})</th>
<th>((\sigma E)_{45^\circ}, \text{MPa})</th>
<th>((F_{tu})_{45^\circ}, \text{MPa})</th>
</tr>
</thead>
<tbody>
<tr>
<td>Longitudinal</td>
<td>1672</td>
<td>0.007908</td>
<td>144.8</td>
<td>220.6</td>
</tr>
<tr>
<td>Transverse</td>
<td>115.3</td>
<td>0.002084</td>
<td>98.6</td>
<td>195.8</td>
</tr>
</tbody>
</table>
Elastic constants.—Average elastic constants were calculated for the ini-
tial linear portions of the stress-strain curves using lamina properties and
laminate theory, and were then compared with the measured values. The average
measured elastic constants for the $[0]_{6T}$ laminate were used for the lamina con-
stants in the analysis. The lamina shear modulus $G_{12}$, which was not measured,
was calculated by

$$G_{12} = \frac{E_y}{2(1 + \nu_{yx})}$$

(15)

where $E_y$ and $\nu_{yx}$ are the constants for the $[\pm 45]_2S$ laminate orientation.

Equation (15) was derived from equation (1) of reference 7. Substituting the
average values from table V into equation (15) and averaging the $G_{12}$ for the
longitudinal and transverse tests gives $G_{12} = 48.68$ GPa. All the calculated
elastic constants are listed in table VI.

Figure 17 shows the calculated constants and the measured elastic con-
stants plotted against the percentage of $45^\circ$ plies for both the longitudinal
and transverse tests. The average and extreme values of the measured constants
are shown. The measured values for the $[\pm 45/0]_S$ and $[0/\pm 45]_S$ laminates show
only a small effect of stacking sequence. The values calculated by laminate
theory are higher than the average measured values by as much as 10 percent for
Young's modulus and by as much as 20 percent for Poisson's ratio. The correla-
tion cannot be improved by changing the value of lamina shear modulus because
Young's modulus for the laminates increases with $G_{12}$.

The ratios of the products $\nu_{xy}E_y$ and $\nu_{yx}E_x$ were calculated from the
average measured elastic constants and the results are shown in the following
table. As anticipated for the orthotropic laminates, the ratios of the prod-
ucts were approximately equal to 1.

<table>
<thead>
<tr>
<th>Laminate orientation</th>
<th>$\nu_{yx}E_x/\nu_{xy}E_y$</th>
</tr>
</thead>
<tbody>
<tr>
<td>$[0]_{6T}$</td>
<td>1.011</td>
</tr>
<tr>
<td>$[0_2/\pm 45]_S$</td>
<td>.986</td>
</tr>
<tr>
<td>$[\pm 45/0_2]_S$</td>
<td>1.001</td>
</tr>
<tr>
<td>$[0/\pm 45]_S$</td>
<td>.986</td>
</tr>
<tr>
<td>$[\pm 45]_2S$</td>
<td>1.024</td>
</tr>
</tbody>
</table>

Note that the linear region of the stress-strain curves, for which the
elastic constants were calculated, does not characterize the laminates over a
significant part of their working range.
Cyclic Tensile Tests

Three loading cycles did not affect the stress and strain at failure for any of the boron/aluminum laminates. The ultimate tensile stress and strain for each cyclic test are presented in table VII, and these values are within the extremes for the static tests given in tables I to V. However, the laminates exhibited permanent strain on unloading. The permanent strains, which increased with applied stress and with the proportion of 45° plies, were significant—especially for the \([45]_{-2S}\) laminate. Figures 18 to 22 show the stress-strain curves for three cycles of loading for one longitudinal and one transverse test of each laminate orientation.

The composite, with the matrix in the "as fabricated" condition, strain-hardened like metals, and the average Ramberg-Osgood curves for monotonic loading form upper envelopes for the cyclic curves. The yield strengths increased dramatically with each load cycle. The hysteresis loops were much more pronounced when some fibers were in the loading direction, probably because the fibers caused the aluminum to yield in compression when unloading from the higher strains. The residual strains were higher for laminates with more 45° plies and were an order of magnitude higher for the \([45]_{-2S}\) laminate than for laminates with 0° plies.

SUMMARY OF RESULTS

The tensile behavior of five types of boron/aluminum laminates \(\left([0]_{6T}, [45]_{2S}, [0/45]_{S}, [02/45]_{S}, \text{ and } [45/02]_{S}\right)\) was investigated. Longitudinal and transverse stress-strain curves were obtained for monotonic loading to failure and for three cycles of loading to successively higher load levels. The Ramberg-Osgood equation was fitted to the stress-strain data to obtain average curves for the various laminate orientations. The following were concluded from the study:

1. For the laminates with 0° plies in the loading direction, the ultimate strengths varied linearly with the percentage of 0° plies in the composite. The strengths predicted by assuming that the 0° plies failed first correlated well with the experimental results.

2. The stress-strain curves for all the laminates were nonlinear except at strains below about 0.00025. In the linear region, measured Young's modulus and Poisson's ratio were within 10 to 20 percent, respectively, of those calculated from laminate theory. However, the linear region does not characterize the laminates over a significant part of their working range.

3. The nonlinearity in the stress-strain curves was greater for higher percentages of 45° plies in the laminate.

4. Laminates without fibers in the loading direction exhibited a very large scatter in the stress and strain at failure.
5. Three cycles of loading did not affect the ultimate stress and strain of the boron/aluminum laminates. The laminates, however, exhibited a permanent strain on unloading. As with the nonlinearity in the stress-strain curves, the amount of permanent strain was greater for laminates with more 45° plies.

6. The Ramberg-Osgood equation fitted the experimental stress-strain curves well.

Langley Research Center
National Aeronautics and Space Administration
Hampton, VA 23665
December 12, 1977
APPENDIX

METHOD FOR FITTING RAMBERG-OSGOOD EQUATION TO STRESS-STRAIN DATA

Equations (7) were fitted to the stress-strain data for longitudinal specimens and equations (8) to the data for transverse specimens. These equations are repeated here for convenience.

\[
\begin{align*}
\varepsilon_y &= \frac{\sigma_y}{E_y} + (b_{yy}\sigma_y)^{nyy} \\
\varepsilon_x &= -\frac{\sigma_{yx}}{E_y} - (b_{yx}\sigma_y)^{nyx} \\
\varepsilon_y &= -\frac{\sigma_{xy}}{E_x} - (b_{xy}\sigma_x)^{nyy} \\
\varepsilon_x &= \frac{\sigma_x}{E_x} + (b_{xx}\sigma_x)^{nxx}
\end{align*}
\]

(A1)

Data from each test were fitted separately using a least-squares method, and the resulting constants were used to define an average curve for each laminate. An iterative least-squares procedure to determine \( E \) (or \( E/\nu \)), \( b \), and \( n \) simultaneously would not converge satisfactorily. Thus, the following two-step iterative procedure was used where \( b \) and \( n \) were determined simultaneously and \( E \) (or \( E/\nu \)) separately.

First, the values of \( E \) and \( E/\nu \) were determined for each test by fitting the linear terms on the right side of equations (A1) and (A2) to only the data in the initial linear region. (Strains in the direction of load in the linear region were less than 0.00025.) A conventional least-squares method was used. Poisson's ratio was then determined from the ratio of \( E \) and \( E/\nu \).

Next, using the calculated values of \( E \) and \( \nu \), the constants \( b \) and \( n \) were determined by fitting equations (A1) and (A2) to all the data from each test. The following iterative, least-squares method (ref. 8, pp. 336-337) was used. The Ramberg-Osgood equation (eq. (6) in the main text),

\[
\varepsilon = \frac{\sigma}{E} + (b\sigma)^n
\]

(A3)

was expanded into a Taylor's series in terms of \( b \) and \( n \), and the higher order terms were dropped. Thus,
APPENDIX

\[ \varepsilon = \varepsilon (b_0, n_0) + \delta b \frac{\partial \varepsilon (b_0, n_0)}{\partial b} + \delta n \frac{\partial \varepsilon (b_0, n_0)}{\partial n} \]  

(A4)

where

\[
\begin{align*}
 b &= b_0 + \delta b \\
n &= n_0 + \delta n
\end{align*}
\]  

(A5)

The \( b_0 \) and \( n_0 \) were taken as approximations to \( b \) and \( n \) and \( \delta b \) and \( \delta n \) as corrections to \( b_0 \) and \( n_0 \). By substituting equation (A3) for \( \varepsilon (b,n) \), equation (A4) becomes

\[ \varepsilon = \frac{\sigma}{E} + (b_0 \sigma)^{n_0} + n_0 \sigma (b_0 \sigma)^{n_0 - 1} \delta b + (b_0 \sigma)^{n_0} (\ln b_0 \sigma) \delta n \]  

(A6)

The corrections \( \delta b \) and \( \delta n \) were calculated to minimize the sum of the squares of the residuals. The residual was the measured strain less the strain calculated by equation (A6).

The approximate values \( b_0 \) and \( n_0 \) were determined initially by passing equation (A3) through two of the data points: one corresponding to the ultimate strain, and the other to one-half of the ultimate strain. The new values of \( b \) and \( n \) calculated by equation (A5) were then used for approximate values to calculate new corrections. This procedure was repeated, generally less than 10 times, until the corrections were less than 1 percent of the values of \( b \) and \( n \). Figure 23 shows examples of experimental stress-strain curves and the fitted Ramberg-Osgood curves. The agreement between these data and equations (A1) and (A2) is typical of all the data.

Figure 24 shows the calculated values of \( b \) and \( n \) plotted as \( \ln b \) against \( n^{-1} \) for the longitudinal tests of the \([0_2/45]_s \) laminate orientation. The lines show significant linear correlation between \( \ln b \) and \( n^{-1} \). All of the other data showed a similar linear correlation. An average Ramberg-Osgood equation for each laminate was defined by using the \( b \) and \( n \) values corresponding to the average values of \( \ln b \) and \( n^{-1} \). Thus, an average behavior was written

\[ \varepsilon = \frac{\sigma}{E} + \frac{(\bar{\ln b})^{(n-1)^{-1}}}{E} \]  

(A7a)

\[ \varepsilon = \frac{\sigma \bar{\nu}}{E} + \frac{(\bar{\ln b})^{(n-1)^{-1}}}{E} \]  

(A7b)

where \( E, \bar{\nu}, \bar{\ln b}, \) and \( n^{-1} \) are the means of \( E, \nu, \ln b, \) and \( n^{-1} \).
APPENDIX

Tables I to V contain the values of $E$, $\nu$, $\ln b$, and $n^{-1}$ for each test and the mean values for each laminate. For the $[0]_6T$ laminate tested transversely, $\varepsilon_y$ was virtually linear with stress $\sigma_x$. For these tests, the iterative procedure did not converge in 20 iterations. Therefore, values of $\ln b_{xy}$ and $(n_{xy})^{-1}$ are not given for these tests in Table I. However, the corrections to $b$ and $n$ always made $b^n$ successively smaller. Thus, the average value of $\ln b_{xy}$ is given as $-\infty$ so that the right-most term in equation (A7b) is zero.
REFERENCES


TABLE I.- STRESS-STRAIN PARAMETERS FOR $[0]_{6T}$ LAMINATE

(a) Longitudinal tests

<table>
<thead>
<tr>
<th>Specimen number</th>
<th>$F_{tu}$, MPa</th>
<th>$\varepsilon_{tu}$</th>
<th>$E_y$, GPa</th>
<th>$\nu_{yx}$</th>
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<th>$(n_{yy})^{-1}$</th>
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(b) Transverse tests

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<th>Specimen number</th>
<th>$F_{tu}$, MPa</th>
<th>$\varepsilon_{tu}$</th>
<th>$E_x$, GPa</th>
<th>$\nu_{xy}$</th>
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<th>$(n_{xx})^{-1}$</th>
<th>$\ln b_{xy}$</th>
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<th>$(n_{xy})^{-1}$ (b)</th>
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$^a$Elastic constants could not be determined from data because $\varepsilon_y$ was not measured below 0.00025. Thus, average elastic constants were assumed.

$^b$Iterative calculation did not converge in 20 iterations; variation of $\varepsilon_y$ with $\sigma_x$ was virtually linear.


TABLE II.- STRESS-STRAIN PARAMETERS FOR \([0_2/\pm 45]_S\) LAMINATE

(a) Longitudinal tests

<table>
<thead>
<tr>
<th>Specimen number</th>
<th>(F_{tu},) MPa</th>
<th>(\varepsilon_{tu})</th>
<th>(E_y,) GPa</th>
<th>(\nu_{yx})</th>
<th>(\ln b_{yy})</th>
<th>((n_{yy})^{-1})</th>
<th>(\ln b_{yx})</th>
<th>((n_{xx})^{-1})</th>
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(b) Transverse tests

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<th>(E_x,) GPa</th>
<th>(\nu_{xy})</th>
<th>(\ln b_{xx})</th>
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TABLE III.- STRESS-STRAIN PARAMETERS FOR \([\pm 45/02]_S\) LAMINATE

(a) Longitudinal tests

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<th>(F_{tu}), MPa</th>
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<th>(E_Y), GPa</th>
<th>(v_{xy})</th>
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(b) Transverse tests

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TABLE IV.- STRESS-STRAIN PARAMETERS FOR \([0/±45]_s\) LAMINATE

(a) Longitudinal tests

<table>
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<th>Specimen number</th>
<th>(F_{tu}), MPa</th>
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(b) Transverse tests

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### TABLE V.- STRESS-STRAIN PARAMETERS FOR $[\pm 45]_{2s}$ LAMINATE

#### (a) Longitudinal tests

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<th>$v_{xy}$</th>
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#### (b) Transverse tests

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<th>$v_{xy}$</th>
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*Strain at which gage failed.

Elastic constants could not be determined from data because $\varepsilon_y$ was not measured below 0.00025. Thus, the average elastic constants from the other tests were used to fit the Ramberg-Osgood equation.
### TABLE VI.- ELASTIC CONSTANTS CALCULATED FROM LAMINATE THEORY

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<th>Laminate orientation</th>
<th>$E_y$, GPa</th>
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<th>$\nu_{xy}$</th>
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<tr>
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TABLE VII.- ULTIMATE TENSILE STRENGTH AND STRAIN FOR CYCLIC TESTS

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<p>| | | | |</p>
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*a Strain at which gage failed.*
Figure 1.- Specimen layouts for the two sizes of boron/aluminum sheets.
Channel numbers | Signal
---|---
1 | Strain gage voltage
2 | Load
3 | $\epsilon_\alpha$ strain (front)
4 | $\epsilon_\alpha$ strain (back)
5 | Load
6 | $\epsilon_\xi$ strain (front)
7 | $\epsilon_\xi$ strain (back)
8 | Load
9 | $\epsilon_\beta$ strain (front)
10 | $\epsilon_\beta$ strain (back)
11 | Load
12 | Peak-load reader

Figure 2. - Recording format for data.
Figure 3. - Schematic diagram of data acquisition system.
Figure 4.- Stress-strain curves for $[0]_{6T}$ laminates.
Experimental curves

Average Ramberg-Osgood curves

(b) Transverse tests.

Figure 4.- Concluded.
Figure 5.- Stress-strain curves for $[0_2/\pm 45]_S$ laminates.
Experimental curves

Average Ramberg-Osgood curves

(b) Transverse tests.

Figure 5.- Concluded.
Experimental curves

Average Ramberg-Osgood curves

Strain

(a) Longitudinal tests.

Figure 6.- Stress-strain curves for $[±{45/0}_2]^S$ laminates.
(b) Transverse tests.

Figure 6. Concluded.
Figure 7. Stress-strain curves for $[0/\pm 45]_s$ laminates.
Experimental curves

Average Ramberg-Osgood curves

(b) Transverse tests.

Figure 7.- Concluded.
Experimental curves

Figure 8.- Stress-strain curves for $[\pm 45]_{2S}$ laminates.
Figure 8.- Concluded.
Figure 9.—Average Ramberg-Osgood stress-strain curves for boron/aluminum laminates.
Average Ramberg-Osgood equation (eq. 11)

(a) Longitudinal tests.

(b) Transverse tests.

Figure 10.- Tangent moduli and Poisson's ratios for $[0]_{6T}$ laminates.
Figure 11.- Tangent moduli and Poisson's ratios for $[0_2/\pm 45]_S$ laminates.
Figure 12. - Tangent moduli and Poisson's ratios for $[\pm 45/0_2]_S$ laminates.
Figure 13.- Tangent moduli and Poisson's ratios for $[0/\pm 45]_S$ laminates.
Figure 14.- Tangent moduli and Poisson's ratios for $[\pm 45]_{2S}$ laminates.
(a) Longitudinal tests of $[0]_{6T}$ laminates.

(b) Transverse tests of $[0]_{6T}$ laminates.

Figure 15. Tangent moduli at small strains.
(c) Longitudinal tests of $[0_2/\pm45]_S$ laminates.

(d) Transverse tests of $[0_2/\pm45]_S$ laminates.

Figure 15.- Continued.
(e) Longitudinal tests of \([+45/0_2]_S\) laminates.

(f) Transverse tests of \([-45/0_2]_S\) laminates.

Figure 15.- Continued.
(g) Longitudinal tests of \([0/\pm 45]_S\) laminates.

(h) Transverse tests of \([0/\pm 45]_S\) laminates.

Figure 15.—Continued.
(i) Longitudinal tests of $[\pm 45]_{2S}$ laminates.

(j) Transverse tests of $[\pm 45]_{2S}$ laminates.

Figure 15.- Concluded.
Failure governed by $0^\circ$ ply failure (eq. (13))

Failure governed by $45^\circ$ ply failure (eq. (14))

Figure 16.- Ultimate tensile strength for longitudinal and transverse tests.
Figure 17.- Elastic constants for longitudinal and transverse tests.

(a) Young's modulus.
Figure 17.- Concluded.
Figure 18.- Stress-strain curve for $[0]_{6T}$ laminate subjected to three loading cycles.
Figure 18.- Concluded.
Figure 19.- Stress-strain curve for $[0_2/\pm 45]_S$ laminate subjected to three loading cycles.
Figure 19.— Concluded.

(b) Transverse test.
Figure 20. Stress-strain curve for $[\pm 45/0_2]_S$ laminate subjected to three loading cycles.
Experimental curves

Average static Ramberg-Osgood curves

Figure 20.- Concluded.

(b) Transverse test.
Figure 21. Stress-strain curve for $[0/±45]_S$ laminate subjected to three loading cycles.

(a) Longitudinal test.
Figure 21.- Concluded.

(b) Transverse test.

- Experimental curves
- Average static Ramberg-Osgood curves
Experimental curves

Average static Ramberg-Osgood curves

Figure 22.- Stress-strain curve for $[\pm 45]_{2S}$ laminate subjected to three loading cycles.

(a) Longitudinal test.
Experimental curves

○ Average static Ramberg-Osgood curves

(b) Transverse test.

Figure 22.- Concluded.
(a) Longitudinal test (specimen no. 2628).

(b) Transverse test (specimen no. 2629).

Figure 23.- Fit of Ramberg-Osgood equation to stress-strain data for a \([0/\pm45]\) specimen.
Figure 24.- Ramberg-Osgood constants for longitudinal tests of \([0_2/\pm 45]_S\) laminates.
The tensile stress-strain behavior of five types of boron/aluminum laminates was investigated. Two of the laminate types contain only 0° or 45° fibers ([0]_6T and [±45]_2S); the remaining three consisted of both 0° and 45° plies ([0/±45], [0/2/±45], and [±45/02]). Longitudinal and transverse stress-strain curves were obtained for monotonic loading to failure and for three cycles of loading to successively higher load levels.

The laminate strengths predicted by assuming that the 0° plies failed first correlated well with the experimental results. The stress-strain curves for all the boron/aluminum laminates were nonlinear except at very small strains. Within the small linear regions, elastic constants calculated from laminate theory corresponded to those obtained experimentally to within 10 to 20 percent. A limited amount of cyclic loading did not affect the ultimate strength and strain for the boron/aluminum laminates. The laminates, however, exhibited a permanent strain on unloading.

The Ramberg-Osgood equation was fitted to the stress-strain curves to obtain average curves for the various laminates. The equation fitted the experimental data well.