OPS MCC
Level B/C
Formulation Requirements

Area Targets and Space Volumes Processor

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OPS MCC LEVEL B/C FORMULATION REQUIREMENTS
AREA TARGETS AND SPACE VOLUMES PROCESSOR

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ACRONYMS AND SYMBOLS

AOS  Acquisition-of-signal
ATSVP  Area Targets and Space Volumes Processor
LOS  Loss-of-signal
MCC  Mission Control Center
M50  Mean-of-1950
RNP  Rotation, nutation, and precession
S/C  Spacecraft
TCA  Time of closest approach
TEI  True-of-epoch inertial

Symbols:

a  Semimajor axis
\(a_F\)  "Adjustment factor" for computing altitude above the Fischer ellipsoid
\(\hat{C}\)  Vector along the centerline of a polyhedron
d  Perpendicular distance to the side of a polyhedron
e  Eccentricity
E  Eccentric anomaly
F  Flattening coefficient
f  True anomaly
h  Altitude
i  Inclination
M  Mean anomaly
n  Number of sides
\(\hat{N}\)  Unit outward normal vector
\( r_c \)  
Radius of the Earth-referenced circle

\( R_{\text{em}} \)  
Mean equatorial radius

\( \mathbf{r}_{\text{sc}} \)  
Spacecraft position vector

\( [\text{RNP}]_{\text{M50}} \)  
RNP matrix from the mean-of-1950 coordinate system to the true-of-epoch inertial system

\( t \)  
Time

\( t_e \)  
Epoch time corresponding to the RNP matrix

\( u \)  
Argument of latitude

\( \alpha \)  
Right ascension

\( \delta \)  
Declination

\( \phi \)  
Geodetic latitude

\( \phi_c \)  
Geocentric latitude

\( \gamma \)  
Central angle.

\( \lambda \)  
Longitude

\( \mu \)  
Earth gravitational constant

\( \mathbf{p} \)  
Slant range vector

\( \tau \)  
Time of perigee passage

\( \omega \)  
Argument of perigee

\( \omega_e \)  
Earth rotation rate

\( \Omega \)  
Right ascension of ascending node

\( \dot{\Omega} \)  
Ascending node vector
Subscripts:
AOS Denotes a parameter associated with the AOS point
B Denotes a vector measured from the lower boundary (i.e., base) of a polyhedron
CA Denotes a parameter associated with the closest approach point
i Denotes the $i^{th}$ vertex or side of a polygon or polyhedron. Also denotes the $i^{th}$ occurrence of an event
LOS Denotes a parameter associated with the LOS point

Superscripts:
G Denotes a parameter in the rotating geocentric coordinate system
TEI Denotes a parameter in the true-of-epoch inertial coordinate system
blank Denotes a parameter in the mean-of-1950 coordinate system
Denotes a unit vector
Denotes a vector
1.0 INTRODUCTION

This document provides the level B/C mathematical specifications for the Area Targets and Space Volumes Processor (ATSVP). Pursuant to the requirements of reference 1, this processor is designed to compute the acquisition-of-signal (AOS) and loss-of-signal (LOS) times for the following:

a. Area targets

(1) Earth-referenced circles which are specified by a latitude, longitude, altitude, and radius.

(2) Celestial circles which are specified by a right ascension, declination, and angular radius.

(3) Earth-referenced polygons which are an arbitrary Earth-fixed figure having up to five sides with the "corner points" defined by latitude, longitude, and altitude.

(4) Celestial polygons which are an arbitrary, inertially fixed figure having up to five sides with the corner points defined by right ascension and declination on the celestial sphere.

b. Space volumes

(1) Earth-referenced space volumes which are an arbitrary, Earth-fixed, five-sided polyhedron. These volumes are defined by a lower-limit polygon at an altitude, \( h_1 \), and the projection of this polygon to an altitude, \( h_2 \). The corner points of the polygon are defined by latitudes and longitudes and rotate with the Earth.

(2) Celestial-fixed space volumes which are an arbitrary, inertially fixed, five-sided polyhedron. These volumes are defined by a lower-limit polygon at an altitude, \( h_1 \), and the projection of this polygon to an altitude, \( h_2 \). The corner points of the polygon are defined by right ascension and declination on the celestial sphere.
The AOS and LOS times for these targets are defined (ref. 1) as follows:

a. Ground circles and polygons
   AOS - the time corresponding to the first subsatellite point to lie just inside the area.
   LOS - the time corresponding to the last subsatellite point just prior to exiting the area.

b. Celestial circles and polygons
   AOS - the time corresponding to the first zenith point to lie just inside the area.
   LOS - the time corresponding to the last zenith point just prior to exiting the area.

c. Earth-referenced and celestial-fixed space volumes
   AOS - the time at which the spacecraft (S/C) is just entering the volume.
   LOS - the time just prior to the S/C exiting the volume.

Six data tables will contain the information necessary to completely describe the area targets and space volumes. These tables (ref. 1) are as follows:

a. Ground targets table containing 10 targets in 1 block of data.
b. Celestial circles table containing 10 targets in 1 block of data.
c. Ground polygons table containing 20 targets in 2 blocks of data.
d. Celestial polygons table containing 10 targets in 1 block of data.
e. Earth-referenced space volumes table containing 10 targets in 1 block of data.
f. Celestial-fixed volumes table containing 10 targets in 1 block of data.

Section 2 of this document describes the characteristics of the area targets and space volumes and provides the mathematical equations necessary to determine whether the S/C lies within the area target or space volume. These equations provide a detailed model of the target geometry and will be used during the precise numerical search.
Section 3 discusses a semianalytical technique for predicting the AOS and LOS time periods. This technique is designed to bound the actual visibility period using a simplified target geometry model and unperturbed orbital motion. Its principal purpose is to reduce the burden on the precise numerical search by eliminating regions of the S/C orbit where AOS and LOS times are physically impossible. Section 4 provides a functional overview of the ATSVP. This section outlines the overall process required to determine precise AOS and LOS times.

Section 5 presents the detailed logic flow for the ATSVP. This section integrates the functional overview presented in section 4 with the equations and approach presented in sections 2 and 3 and the appendixes. Appendix A discusses the procedure for subdividing complex concave polygons into two or more simpler convex segments. The purpose of this subdivision process is to permit the equations in section 2 to be used on a segment-by-segment basis to test for containment. Appendix B provides a solution to the conic intersection equations used in section 3 for celestial-fixed targets.
2.0 AREA TARGETS AND SPACE VOLUMES CHARACTERISTICS AND CONTAINMENT CRITERIA

The following subsections discuss the characteristics of each of the area targets and space volumes presented in section 1. The mathematical equations necessary to determine whether the S/C lies within the area target or space volume are also developed and discussed.

Two reference coordinate systems will be used. The inertial Aries mean-of-1950 (M50) coordinate system (fig. 2-1) will be the reference system when dealing with area targets and space volumes which remain inertially fixed. The rotating geocentric coordinate system (fig. 2-2) will be used when dealing with area targets and space volumes which rotate with the Earth.

Each of the following six subsections is further subdivided into three topics:

a. Procedure - a brief description of the steps to be performed.
b. Equations - a statement of the input parameter requirements and development of the mathematical equations.
c. Assumptions and limitations - a description of any simplifying assumptions and/or mathematical restrictions.

For convenience, section 2.7 summarizes the equations for all of the area targets and space volumes.
NAME: Aries mean-of-1950, Cartesian, coordinate system.

ORIGIN: The center of the Earth.

ORIENTATION: The epoch is the beginning of Besselian year 1950 or Julian ephemeris date 2433282.423357.

The $X_M$-$Y_M$ plane is the mean Earth's equator of epoch.

The $X_M$ axis is directed towards the mean vernal equinox of epoch.

The $Z_M$ axis is directed along the Earth's mean rotational axis of epoch and is positive north.

The $Y_M$ axis completes a right-handed system.

CHARACTERISTICS: Inertial, right-handed, Cartesian system.

---

Figure 2-1.- Aries mean-of-1950 coordinate system.
NAME: Geocentric coordinate system

ORIGIN: Center of the Earth

ORIENTATION: $X_G - Y_G$ plane is the Earth's true-of-date equator
$X_G$ passes through the Greenwich meridian
$Z_G$ is along the Earth's rotational axis
$Y_G$ completes the right-handed system

CHARACTERISTICS: Rotating, right-handed, Earth-fixed

Figure 2-2.- Rotating geocentric coordinate system.
2.1 EARTH-REFERENCED CIRCLES

Earth-referenced circles are defined to be circular ground target areas whose centers are defined by geodetic latitude, longitude, and altitude (fig. 2-3). The S/C lies within this ground target area if its subsatellite point lies within the perimeter of the circular area.

2.1.1 Procedure

The following procedure will be used to determine whether the S/C lies within the ground target area:

a. The geodetic coordinates of the ground target area will be transformed to the geocentric system.

b. The S/C position vector will be transformed from the M50 system to the geocentric system.

c. A test will be performed to determine whether the S/C subsatellite point lies within the perimeter of the ground target area.

2.1.2 Equations

The following parameters are required:

\( \phi \) and \( \lambda \) - geodetic latitude and longitude, respectively, of the center of the Earth-referenced circle (fig. 2-4)

\( h \) - altitude of the Earth-referenced circle, measured with respect to the Fischer ellipsoid of 1960 (fig. 2-4)

\( r_c \) - radius of the Earth-referenced circle (fig. 2-3)

\( \vec{r}_{sc} \) - S/C position vector in the M50 system (fig. 2-3)

\( t \) - time corresponding to \( \vec{r}_{sc} \)

\([\text{RNP}]_{\text{M50}}^{\text{TEI}}\) - Rotation, nutation, and precession (RNP) matrix which is used to transform vectors from the M50 system to the true-of-epoch inertial (TEI) coordinate system

\( t_e \) - epoch time corresponding to the RNP matrix

\( R_{em} \) - mean equatorial radius for the Fischer ellipsoid of 1960
Figure 2-3. - Earth-referenced circle.
NAME: 	Geodetic coordinate system.

ORIGIN: 	This system consists of a set of parameters rather than a coordinate system; therefore, no origin is specified.

ORIENTATION: 	This system of parameters is based on an ellipsoidal model of the Earth (e.g., the Fischer ellipse of 1960). For any point of interest we define a line, known as the geodetic local vertical, which is perpendicular to the ellipsoid and which contains the point of interest.

$h$, geodetic altitude, is the distance from the point of interest to the reference ellipsoid, measured along the geodetic local vertical, and is positive for points outside the ellipsoid.

$\lambda$ is the longitude measured in the plane of the Earth's true equator from the prime (Greenwich) meridian to the local meridian, measured positive eastward.

$\phi$ is the geodetic latitude, measured in the plane of the local meridian from the Earth's true equator to the geodetic local vertical, measured positive north from the equator.

NOTE: A detailed explanation of declination, geodetic latitude, and geocentric latitude is provided on figure 2-4(b).

CHARACTERISTICS: Rotating polar coordinate parameters. Only position vectors are expressed in this coordinate system. Velocity vectors should be expressed in the Aries mean-of-1950, or the Aries true-of-date, polar for inertial or quasi-inertial representations, respectively. The Fischer ellipsoid model should be used with this system.

(a) Basic definitions.

Figure 2-4.- Geodetic coordinate system.

2-6
NAME: Geodetic coordinate system of point P.

DEFINITIONS:

h is the altitude of point P measured perpendicular from the surface of the referenced ellipsoid.

ϕ is the geodetic latitude of point P.

ϕ̄ is the geocentric latitude of point P.

δ is the angle between radius vector and equatorial plane (declination).

λ is the longitude of point P. Angle (+ east) between plane of the figure and the plane formed by the Greenwich meridian.

(b) Detailed explanation.

Figure 2-4.-- Concluded.
\( F \) - flattening coefficient for the Fischer ellipsoid of 1960

\( \omega_e \) - Earth rotation rate

The first step is to transform the ground target area from the geodetic system to the geocentric system. The vector from the center of the Earth to the center of the ground target area can be expressed in the rotating geocentric system by

\[
C^G_B = \begin{bmatrix}
(h + a_F) \cos \lambda \cos \phi \\
(h + a_F^0) \sin \lambda \cos \phi \\
[h + (1 - F)^2 a_F^0] \sin \phi
\end{bmatrix}
\]  

(2-1)

where

\[
a_F = \frac{R_{em}}{\sqrt{\cos^2 \phi + (1 - F)^2 \sin^2 \phi}}
\]  

(2-2)

The second step is to transform the S/C position vector from the M50 system to the geocentric system. This is accomplished by

a. Transforming the vector from the M50 system to the TEI system using the RNP matrix.

b. Transforming the resultant vector from the TEI system to the geocentric system.

The S/C position vector in the TEI system is given by

\[
R^{TEI}_{sc} = [RNP]^{TEI}_{M50} R^{\text{TEI}}_{sc}
\]  

(2-3)

The S/C position vector in the geocentric system (fig. 2-5) is given by

2-8
Figure 2-5.- Relationship between TEI and rotating geocentric systems.
\[ \begin{pmatrix} \cos \Delta \lambda & \sin \Delta \lambda & 0 \\ -\sin \Delta \lambda & \cos \Delta \lambda & 0 \\ 0 & 0 & 1 \end{pmatrix} \]

where

\[ \Delta \lambda = \omega (t - t_e) \]  \hspace{1cm} (2-5)

The final step is to determine whether the S/C subsatellite point lies within the ground target area (fig. 2-3). The angle from \( \mathcal{C}_B^G \) to the perimeter of the circular ground target area, \( \gamma_A \), is given (in degrees) by

\[ \gamma_A = \frac{r_C}{\mathcal{C}_B^G} \left( \frac{180}{\pi} \right) \]  \hspace{1cm} (2-6)

The angle from \( \mathcal{C}_B^G \) to the S/C, \( \gamma_S \), is given by

\[ \gamma_S = \cos^{-1} \left( \frac{\mathcal{C}_B^G \cdot \mathcal{C}_B}{|\mathcal{C}_B^G| \cdot |\mathcal{C}_B|} \right) \]  \hspace{1cm} (2-7)

The S/C lies within the ground target area if

\[ \gamma_S \leq \gamma_A \]  \hspace{1cm} (2-8)

2.1.3 Assumptions and Limitations

The following assumptions are implicit in the equations presented in section 2.1.2:

a. The S/C geocentric subsatellite point is used to compute entry into the ground target area.

b. The effects of polar nutation and precession from time \( t_e \) to time \( t \) can be neglected.
c. The radius of the Earth-referenced circle, \( r_c \), is a segment of arc.
2.2 CELESTIAL CIRCLES

Celestial circles are defined to be circular areas on the celestial sphere (fig. 2-6). The center of this area target is defined by right ascension and declination. The criterion for a S/C to lie within this area is for the S/C zenith point to lie within the perimeter of the celestial circle.

2.2.1 Procedure

The following procedure will be used to determine whether the S/C lies within the celestial circle:

a. The unit vector along the centerline of the celestial circle will be computed in the M50 system.

b. The dot product between this vector and the S/C position vector will be formed to determine whether the S/C lies within the celestial circle.

2.2.2 Equations

The following parameters are required:

\( \alpha_A \) and \( \delta_A \) - the right ascension and declination, respectively, of the centerline of the celestial circle expressed in the M50 system (fig. 2-6).

\( \gamma_A \) - the celestial circle angular radius (fig. 2-6).

\( \mathbf{R}_{sc} \) - S/C position vector in the M50 system.

The unit vector from the center of the Earth to the center of the celestial circle, \( \hat{\mathbf{C}}_B \), is given by

\[
\hat{\mathbf{C}}_B = \begin{pmatrix}
\cos \alpha_A \cos \delta_A \\
\sin \alpha_A \cos \delta_A \\
\sin \delta_A 
\end{pmatrix}
\]  

(2-9)
Figure 2-6.- Containment test for celestial circles.
The angle between this vector and the S/C, $\gamma_S$, is

$$\gamma_S = \cos^{-1} \left( \hat{C}_B \cdot \left| \frac{\hat{R}_{SC}}{|\hat{R}_{SC}|} \right| \right) \quad (2-10)$$

The S/C lies within the celestial circle if

$$\gamma_S \leq \gamma_A \quad (2-11)$$

2.2.3 **Assumptions and Limitations**

None.
2.3 EARTH-REFERENCED POLYGONS

The Earth-referenced polygon is defined to be an arbitrary planar figure having up to five sides (fig. 2-7). This figure is fixed with respect to the rotating Earth. The corner points (i.e., vertices) of this polygon are defined by geodetic latitude, longitude, and altitude. The basic criterion for penetration is to ensure that the S/C subsatellite point lies within the perimeter of the ground target area.

2.3.1 Procedure

The basic procedure for this ground target area is similar to the procedure presented in section 2.1.1. It consists of

a. Transforming the geodetic coordinates of each polygon vertex to the geocentric system.

b. Transforming the S/C position vector from the M50 system to the geocentric system.

c. Testing to determine whether the S/C is interior to all planes defined by the sides of the polygon.

2.3.2 Equations

The following parameters are required:

- \( n \) - number of sides
- \( \phi_i \) and \( \lambda_i \) - geodetic latitude and longitude, respectively, of each vertex (fig. 2-7)
- \( h_i \) - geodetic altitude of each vertex (fig. 2-7)
- \( \mathbf{R}_{sc} \) - S/C position vector in the M50 system
- \( t \) - time corresponding to \( \mathbf{R}_{sc} \)
- \( [\text{RNP}]_{M50}^{\text{TEI}} \) - RNP matrix to transform from the M50 system to the TEI system
- \( t_e \) - epoch time corresponding to the RNP matrix
- \( R_{em} \) - mean equatorial radius for the Fischer ellipsoid of 1960
- \( F \) - flattening coefficient for the Fischer ellipsoid of 1960
- \( \omega_e \) - Earth rotation rate
Fischer (1960) ellipsoid

True-of-date equator

Greenwich meridian

Figure 2-7.- Earth-referenced polygon.
The first step is to transform the parameters defining each vertex of the polygon from the geodetic system to the rotating geocentric system. These vectors are given by

\[
\hat{\mathbf{R}}_i^G = \begin{cases} 
(h_i + a_F) \cos \lambda_i \cos \phi_i \\
(h_i + a_F) \sin \lambda_i \cos \phi_i \\
[h_i + (1 - F)^2 a_F] \sin \phi_i 
\end{cases} \quad i = 1, 2, 3, \ldots n
\]

(2-12)

where

\( a_F \) is defined by equation 2-2 with \( \phi_i \) replacing \( \phi \).

The S/C position vector in the geocentric system, \( \hat{\mathbf{R}}_{sc}^G \), is then obtained by using equations 2-3 through 2-5.

The next step in the procedure is to determine whether the S/C lies interior to all planes defined by the sides of the polygon. Figures 2-8 and 2-9 illustrate the geometry. All vectors in these figures are with respect to the geocentric system. The centroid of the ground target area, \( \mathbf{c}_B^G \) (fig. 2-8) is defined as

\[
\mathbf{c}_B^G = \frac{\sum_{i=1}^{n} \hat{\mathbf{R}}_i^G}{n}
\]

(2-13)

The unit normal vectors to each side of the polygon (fig. 2-9) are given by

\[
\hat{\mathbf{N}}_i^G = \frac{\hat{\mathbf{R}}_{i+1}^G \times \hat{\mathbf{R}}_i^G}{|\hat{\mathbf{R}}_{i+1}^G \times \hat{\mathbf{R}}_i^G|} \quad i = 1, 2, 3, \ldots n-1
\]

(2-14a)

\[
\hat{\mathbf{N}}_n^G = \frac{\hat{\mathbf{R}}_1^G \times \hat{\mathbf{R}}_n^G}{|\hat{\mathbf{R}}_1^G \times \hat{\mathbf{R}}_n^G|}
\]

(2-14b)
Figure 2-8.- Centroid of Earth-referenced polygons.
Figure 2-9.- Containment test for Earth-referenced polygons.
The perpendicular distance from $\mathbf{c}^G_B$ to each polygon side is

$$d_i = N_i \cdot \left( \mathbf{r}^G_i - \mathbf{c}^G_B \right), \quad i = 1, 2, 3, \ldots n \quad (2-15)$$

The S/C must lie interior to the $i^{th}$ plane if

$$\nabla N_i \cdot \mathbf{\rho}_B \leq d_i \quad (2-16)$$

where

$$\mathbf{\rho}_B = \mathbf{r}_{sc} - \mathbf{c}^G_B \quad (2-17)$$

If equation 2-16 is satisfied for all sides, then the S/C subsatellite point lies within the perimeter of the polygon.

2.3.3 Assumptions and Limitations

The following assumptions and limitations are implicit in the equations presented in section 2.3.2:

a. The S/C geocentric subsatellite point is used to determine entry into the ground target area.

b. The effects of polar nutation and precession, from time $t_e$ to $t$ can be neglected.

c. The vertices of the polygon are specified in a counterclockwise order as viewed from the top.

d. The polygon is convex; i.e., the interior angles between the sides defining the vertices are less than 180 degrees.\(^1\)

e. The angular separation between two consecutive vertices is sufficient to permit a nonzero cross product (eq. 2-14).

f. The S/C and the ground target area lie in the same hemisphere. The procedure discussed in section 3 will assure this condition.

\(^1\)Concave polygons can be accommodated by subdividing them into two or more convex polygons. Appendix A discusses this procedure.
2.4 CELESTIAL POLYGONS

Celestial polygons are defined to be arbitrary figures having up to five sides with the corner points (i.e., vertices) defined by right ascension and declination on the celestial sphere. The criterion for penetration into this area is to ensure that the S/C zenith point lies within the confines of the polygon. Figure 2-10 illustrates the celestial polygon. It is noted that this polygon also represents a variable area polyhedron which remains inertially fixed.

2.4.1 Procedure

The basic procedure for this area target is similar to the procedure presented in section 2.3.1, i.e.,:

a. The unit vector along the "centroid" of the polyhedron will be computed in the M50 system.

b. Tests will be made to determine whether the S/C is interior to all planes defined by the sides of the polyhedron.

2.4.2 Equations

The following parameters are required:

- \( n \) - number of sides
- \( \alpha_i \) and \( \delta_i \) - right ascension and declination, respectively, of each vertex in the M50 system (i = 1,2,3,...n)
- \( \hat{R}_{SC} \) - S/C position vector in the M50 system

The first step is to determine the unit vector along the centroid of the polyhedron. The unit vectors from the center of the Earth to each vertex in the M50 system are given by

\[
\hat{R}_i = \begin{pmatrix}
\cos \alpha_i \cos \delta_i \\
\sin \alpha_i \cos \delta_i \\
\sin \delta_i
\end{pmatrix} \quad i = 1,2,3,...n \quad (2-18)
\]
Figure 2-10.- Celestial polygon.
The unit vector along the centroid of the polyhedron is

\[ \hat{C} = \frac{\hat{r}_i}{|\hat{r}_i|} \quad (2-19) \]

where

\[ \hat{C} = \frac{\sum_{i=1}^{n} \hat{r}_i}{n} \quad (2-20) \]

The final step is to determine whether the S/C lies interior to all planes defined by the sides of the polyhedron. Figure 2-11 illustrates the geometry. The slant range from \( \hat{C} \) to the S/C is

\[ \hat{\rho}_B = \hat{r}_{SC} - \hat{C} \quad (2-21) \]

Equations 2-14 through 2-16 (with \( \hat{r}^G_i \), \( \hat{c}^G \), and \( \hat{\rho}^G \) replaced by \( \hat{r}_i \), \( \hat{C} \), and \( \hat{\rho}_B \), respectively) are then used to determine whether the S/C is interior to all polyhedron planes.

2.4.3 Assumptions and Limitations

The following assumptions and limitations are implicit in the equations presented in section 2.4.2:

a. The vertices of the polygon are specified in a counterclockwise order as viewed from the top.

b. The polygon is convex; i.e., the interior angles between the sides defining the vertices are less than 180 degrees.\(^1\)

c. The angular separation between two consecutive vertices is sufficient to permit a nonzero cross product (eq. 2-14).

d. The S/C and the area target lie within the same hemisphere. The procedure outlined in section 3 will assure this condition.

\(^1\) Concave polygons can be accommodated by subdividing them into two or more convex polygons as discussed in appendix A.
Figure 2-11.- Containment test for celestial polygons.
2.5 EARTH-REFERENCED SPACE VOLUMES

The Earth-referenced space volumes are arbitrary five-sided polyhedrons which are defined by a lower-limit polygon at an altitude, $h_1$, and the projection of this polygon to an altitude, $h_2$. The vertices of this polygon are defined by geodetic latitude and longitude and rotate with the Earth. Figure 2-12 illustrates this type of space volume. As shown, the planar cross sectional area of this polyhedron remains constant with respect to altitude.

2.5.1 Procedure

The following procedure will be used to determine whether the S/C lies within the space volume:

a. The geodetic parameters defining each vertex of the lower boundary will be transformed to geocentric position vectors.

b. The centroid vector to the lower boundary will be computed in the geocentric system.

c. The S/C position vector will be transformed from the M50 system to the geocentric system.

d. Tests will be performed to ensure that the S/C lies above the lower boundary and below the upper boundary. If either of these tests fails, then no further computations are required.

e. Assuming the previous step is passed, tests will be performed to determine whether the S/C is interior to all planes defined by the sides of the polyhedron.

2.5.2 Equations

The following parameters are required:

$\phi_i$ and $\lambda_i$ - geodetic latitude and longitude, respectively, of each vertex of the lower boundary ($i = 1, 2, 3, 4, 5$)

$h_1$ - geodetic altitude of the lower boundary

$h_2$ - geodetic altitude of the upper boundary
Figure 2-12.- Constant area polyhedron.
$\hat{R}_{sc}$ - S/C position vector in the M50 system
$t_{TEI}$ - time corresponding to $\hat{R}_{sc}$
$[\text{RNP}]_{M50}^{\text{TEI}}$ - RNP matrix to transform from the M50 to TEI system
t_e - epoch time corresponding to the RNP matrix
$R_{em}$ - mean equatorial radius for the Fischer ellipsoid of 1960
$F$ - flattening coefficient for the Fischer ellipsoid of 1960
$\omega_e$ - Earth rotation rate

The first step is to transform the parameters defining each vertex of the lower boundary to geocentric position vectors. Equation 2-12 (with $h_1 = 0$) provides the necessary transformation.

The centroid of these vectors is given by

$$\hat{r}_G = \frac{\sum_{i=1}^{n} \hat{r}_G^i}{n} \quad (2-22)$$

The geocentric vector to the centroid of the lower boundary (fig. 2-13) is given by

$$\hat{r}_B^G = \hat{r}_G + h_1 \frac{\hat{r}_G}{|\hat{r}_G|} \quad (2-23)$$

Next, the S/C position vector in the geocentric system, $\hat{r}_{sc}^G$, is computed using equations 2-3 through 2-5.

The fourth step in the procedure is to ensure that the S/C lies between the upper and lower boundaries. Figure 2-13 illustrates the geometry. All vectors in this figure are with respect to the geocentric system. The slant range from $\hat{r}_B^G$ to the S/C, $\hat{r}_B^c$,
Center of Earth

Figure 2-13.- Boundary tests for constant area polyhedrons.
is given by

$$\hat{C}_B = R_{SC} - \hat{C}_B$$  \hspace{2cm} (2-24)

The S/C lies above the lower boundary and below the upper boundary if

$$h_2 - h_1 \geq \rho_B \cdot \frac{\hat{C}_B}{|\hat{C}_B|} \geq 0$$  \hspace{2cm} (2-25)

Assuming that equation 2-25 is satisfied, the final step is to determine whether the S/C lies interior to the side planes of the polyhedron. Figure 2-14 illustrates the geometry. The unit normal vectors from \( \hat{C}_B \) to each side of the polygon are given by

$$\hat{N}_i = \frac{\hat{C}_B \times \left( \hat{C}_B \times \left( \hat{C}_B \times \left( \hat{C}_B \times \left( \hat{C}_B \times \hat{C}_B \right) \right) \right) \right)}{||\hat{C}_B \times \left( \hat{C}_B \times \left( \hat{C}_B \times \left( \hat{C}_B \times \hat{C}_B \right) \right) \right)||} \quad \text{for} \quad i = 1,2,3,4$$  \hspace{2cm} (2-26a)

$$\hat{N}_s = \frac{\hat{C}_B \times \left( \hat{C}_B \times \left( \hat{C}_B \times \left( \hat{C}_B \times \hat{C}_B \right) \right) \right)}{||\hat{C}_B \times \left( \hat{C}_B \times \left( \hat{C}_B \times \left( \hat{C}_B \times \hat{C}_B \right) \right) \right)||} \quad \text{for} \quad s = 5,6$$  \hspace{2cm} (2-26b)

Equations 2-15 and 2-16 are then used to determine whether the S/C is contained within the polyhedron.

### 2.5.3 Assumptions and Limitations

The following assumptions and limitations are implicit in the equations presented in section 2.5.2:

a. The altitude of the lower boundary, \( h_1 \), is measured with respect to the centroid of the vertices which lie on the Fischer ellipsoid.
Figure 2-14.— Containment test for constant area polyhedrons.
b. The effects of polar nutation and precession from time $t_e$ to $t$ can be neglected.

c. The vertices are specified in a counterclockwise order as viewed from the top.

d. The planar area of the polyhedron is convex; i.e., the interior angles between the sides defining the vertices are less than 180 degrees.\(^1\)

e. The local horizon for determining whether the S/C is between the upper and lower boundaries is perpendicular to the centroid vector.

f. The distance between two consecutive vertices is sufficient to permit a nonzero cross product (eq. 2-26).

\(^1\)Concave polyhedrons can be accommodated by subdividing them into two or more convex portions as discussed in appendix A.
2.6 CELESTIAL-FIXED SPACE VOLUMES

The celestial-fixed space volumes are arbitrary five-sided polyhedrons which are defined by a lower-limit polygon at an altitude, \( h_1 \), and a projection of this polygon to an altitude, \( h_2 \). The vertices of the polygon are defined by right ascension and declination and are inertially fixed. Figure 2-12 can also be used to illustrate this type of space volume. As mentioned previously, the planar cross sectional area of this polyhedron remains constant with respect to altitude.

2.6.1 Procedure

The basic procedure for this space volume is very similar to the procedure presented in section 2.5.1. It consists of:

a. Computing the centroid vector to the lower boundary in the M50 system.

b. Testing to ensure that the S/C lies above the lower boundary and below the upper boundary. If either of these tests fails, then no further computations are required.

c. Assuming the previous step is passed, further tests will be performed to determine whether the S/C is interior to all planes defined by the sides of the polyhedron.

2.6.2 Equations

The following parameters are required:

\[ \alpha_i \text{ and } \delta_i \] - right ascension and declination, respectively, of each vertex of the lower boundary in the M50 system.

\( h_1 \) - altitude of the lower boundary

\( h_2 \) - altitude of the upper boundary

\( R_{\text{em}} \) - mean equatorial radius

2-32
The unit vector to the centroid of the polyhedron, \( \mathbf{\hat{C}} \), is computed via equations 2-18 through 2-20. The vector to the centroid of the lower boundary is given by

\[
\mathbf{\hat{C}}_B = (\mathbf{R}_{em} + h_1) \mathbf{\hat{C}}
\]  

(2-27)

Equations 2-24 and 2-25 (with variables \( \mathbf{R}^G_{SC} \) and \( \mathbf{C}^G_B \) replaced by \( \mathbf{R}^G_{Sc} \) and \( \mathbf{C}^G_B \), respectively) are then used to ensure that the S/C lies between the upper and lower boundaries.

Assuming equation 2-25 is satisfied, equation 2-26 (with \( \mathbf{C}^G_B \) and \( \mathbf{R}^G_i \) replaced by \( \mathbf{C}_B \) and \( \mathbf{R}_i \), respectively) is used to define the unit normal vectors to each side of the polyhedron. Finally, equations 2-15 and 2-16 are used to determine whether the S/C lies within the space volume.

2.6.3 Assumptions and Limitations

The following assumptions and limitations are implicit in the equations presented in section 2.6.2:

a. The altitudes of the lower and upper boundaries are measured with respect to the mean equatorial radius.

b. The vertices are specified in a counterclockwise order as viewed from the top.

c. The planar area of the polyhedron is convex; i.e., the interior angles between the sides defining the vertices are less than 180 degrees.\(^1\)

d. The distance between two consecutive vertices is sufficient to permit a nonzero cross product (eq. 2-26).

\(^1\) Concave polyhedrons can be accommodated by subdividing them into two or more convex portions as discussed in appendix A.
2.7 SUMMARY OF EQUATIONS

This section summarizes all of the equations presented in the previous sections for the various area targets and space volumes. The order of presentation and equation numbers correspond to the computation sequence discussed in the text.

2.7.1 Earth-Referenced Circles

\[
\begin{align*}
\mathbf{a}_G &= \begin{bmatrix} (h + a_F) \cos \lambda \cos \phi \\ (h + a_F) \sin \lambda \cos \phi \\ h + (1 - F)^2 a_F \sin \phi \end{bmatrix} \\
\mathbf{a}_B &= \mathbf{a}_G \mathbf{R}_{em} \\
\mathbf{a}_F &= \frac{\mathbf{R}_{em}}{\sqrt{\cos^2 \phi + (1 - F)^2 \sin^2 \phi}} \\
\mathbf{R}_{sc}^{TEI} &= [\mathbf{RNP}]^{TEI} \mathbf{R}_{sc} \\
\mathbf{R}_{sc}^G &= \begin{bmatrix} \cos \Delta \lambda & \sin \Delta \lambda & 0 \\ -\sin \Delta \lambda & \cos \Delta \lambda & 0 \\ 0 & 0 & 1 \end{bmatrix} \mathbf{R}_{sc}^{TEI}
\end{align*}
\]

where

\[
\Delta \lambda = \omega_e (t - t_e)
\]
\[ \gamma_A = \left\{ \frac{r_c}{|\mathbf{G}_B|} \right\} \cdot \frac{180}{\pi} \quad (2-6) \]

\[ \gamma_S = \cos^{-1} \left\{ \frac{\mathbf{G}_S \cdot \mathbf{G}_B}{|\mathbf{G}_S| \cdot |\mathbf{G}_B|} \right\} \quad (2-7) \]

\[ \gamma_S \leq \gamma_A \quad (2-8) \]

### 2.7.2 Celestial Circles

\[ \mathbf{C}_B = \begin{Bmatrix} \cos \alpha_A \cos \delta_A \\ \sin \alpha_A \cos \delta_A \\ \sin \delta_A \end{Bmatrix} \quad (2-9) \]

\[ \gamma_S = \cos^{-1} \left\{ \mathbf{C}_B \cdot \frac{\mathbf{G}_S}{|\mathbf{G}_S|} \right\} \quad (2-10) \]

\[ \gamma_S \leq \gamma_A \quad (2-11) \]

### 2.7.3 Earth-Referenced Polygons

\[ \mathbf{R}_i^G = \begin{Bmatrix} (h_i + a_F) \cos \lambda_i \cos \phi_i \\ (h_i + a_F) \sin \lambda_i \cos \phi_i \\ [h_i + (1 - F)^2 a_F] \sin \phi_i \end{Bmatrix} \quad i = 1, 2, 3, \ldots n \quad (2-12) \]
where

\[ a_F = \frac{R_{em}}{\sqrt{\cos^2 \phi_i + (1 - F)^2 \sin^2 \phi_i}} \quad (2-2) \]

\[ \tilde{R}_TEI = [RNP]_{TEI} \tilde{R}_sc \quad (2-3) \]

\[ \tilde{R}_sc^G = \begin{bmatrix} \cos \Delta \lambda & \sin \Delta \lambda & 0 & \tilde{R}^G_{TEI} \\ -\sin \Delta \lambda & \cos \Delta \lambda & 0 \\ 0 & 0 & 1 \end{bmatrix} \quad (2-4) \]

where

\[ \Delta \lambda = \omega_e (t - t_e) \quad (2-5) \]

\[ \tilde{\hat{C}}_B = \frac{\sum_{i=1}^{n} \tilde{R}^G_i}{n} \quad (2-13) \]

\[ \tilde{\hat{N}}_i = \frac{\tilde{R}^C_{i+1}}{|\tilde{R}^C_{i+1} \times \tilde{R}^C_i|} \quad i = 1, 2, 3, \ldots, n-1 \quad (2-14a) \]

\[ \tilde{\hat{N}}_n = \frac{\tilde{R}^C_1 \times \tilde{R}^C_n}{|\tilde{R}^C_1 \times \tilde{R}^C_n|} \quad (2-14b) \]

\[ d_i = \tilde{\hat{N}}_i \cdot (\tilde{R}^G_i - \tilde{\hat{C}}_B) \quad i = 1, 2, 3, \ldots, n \quad (2-15) \]

\[ \tilde{\hat{N}}_i \cdot \tilde{\hat{C}}_B \leq d_i \quad i = 1, 2, 3, \ldots, n \quad (2-16) \]
where

\[ \hat{\rho}_B = \hat{\rho}_{sc} - \hat{C} \]  \hspace{1cm} (2-17)

### 2.7.4 Celestial Polygons

\[ \hat{R}_i = \begin{pmatrix} \cos \alpha_i & \cos \delta_i \\ \sin \alpha_i & \cos \delta_i \\ \sin \delta_i \end{pmatrix} \hspace{1cm} i = 1, 2, 3, \ldots n \]  \hspace{1cm} (2-18)

\[ \hat{C} = \frac{\hat{C}}{|\hat{C}|} \]  \hspace{1cm} (2-19)

where

\[ \hat{C} = \frac{\sum_{i=1}^{n} \hat{R}_i}{n} \]  \hspace{1cm} (2-20)

\[ \hat{\rho}_B = \hat{\rho}_{sc} - \hat{C} \]  \hspace{1cm} (2-21)

\[ \hat{N}_i = \frac{\hat{R}_{i+1} \times \hat{R}_i}{|\hat{R}_{i+1} \times \hat{R}_i|} \hspace{1cm} i = 1, 2, 3, \ldots n-1 \]  \hspace{1cm} (2-14a)

\[ \hat{N}_n = \frac{\hat{R}_1 \times \hat{R}_n}{|\hat{R}_1 \times \hat{R}_n|} \]  \hspace{1cm} (2-14b)

\[ d_i = \hat{N}_i \cdot (\hat{R}_i - \hat{C}) \hspace{1cm} i = 1, 2, 3, \ldots n \]  \hspace{1cm} (2-15)

\[ \hat{N}_i \cdot \hat{\rho}_B \leq d_i \hspace{1cm} i = 1, 2, 3, \ldots n \]  \hspace{1cm} (2-16)
2.7.5 Earth-Referenced Space Volumes

\[ \hat{\mathbf{R}}_i^G = \begin{cases} \\ a_F \cos \lambda_i \cos \phi_i \\ a_F \sin \lambda_i \cos \phi_i \\ (1 - F)^2 a_F \sin \phi_i \end{cases} \quad i = 1, 2, 3, 4, 5 (2-12) \]

where

\[ a_F = \frac{R_{em}}{\sqrt{\cos^2 \phi_i + (1 - F)^2 \sin^2 \phi_i}} \quad (2-2) \]

\[ \tilde{c}_G = \frac{\sum_{i=1}^{5} \hat{R}_i^G}{5} \quad (2-22) \]

\[ \tilde{c}_B^G = \tilde{c}_G + h_1 \frac{\tilde{c}_G}{|\tilde{c}_G|} \quad (2-23) \]

\[ \hat{R}_{sc}^{TEI} = [RNP]_{TEI} \hat{R}_{M50} \quad (2-3) \]

\[ \hat{R}_{sc}^G = \begin{bmatrix} \cos \Delta \lambda & \sin \Delta \lambda & 0 \\ -\sin \Delta \lambda & \cos \Delta \lambda & 0 \\ 0 & 0 & 1 \end{bmatrix} \hat{R}_{sc}^{TEI} \quad (2-4) \]

where

\[ \Delta \lambda = \omega_e (t - t_e) \quad (2-5) \]

\[ \hat{R}_B^G = \hat{R}_{sc} - \tilde{c}_B \quad (2-24) \]
\[ h_2 - h_1 \geq \hat{\rho}_B \cdot \frac{\hat{c}_G}{|\hat{c}_B|} \geq 0 \quad (2-25) \]

\[ \hat{N}_i^G = \frac{\hat{c}_G \times \left( \hat{R}_i^G - \hat{R}_{i+1}^G \right)}{|\hat{c}_B \times \left( \hat{R}_i^G - \hat{R}_{i+1}^G \right)|} \quad i = 1, 2, 3, 4 \quad (2-26a) \]

\[ \hat{N}_5^G = \frac{\hat{c}_G \times \left( \hat{R}_5^G - \hat{R}_1^G \right)}{|\hat{c}_B \times \left( \hat{R}_5^G - \hat{R}_1^G \right)|} \quad (2-26b) \]

\[ l_i = \hat{N}_i^G \cdot \left( \hat{R}_i^G - \hat{c}_B \right) \quad i = 1, 2, 3, 4, 5 \quad (2-15) \]

\[ \hat{N}_i^G \cdot \hat{\rho}_B \leq d_i \quad i = 1, 2, 3, 4, 5 \quad (2-16) \]

2.7.6 Celestial-Fixed Space Volumes

\[ \hat{R}_i = \begin{pmatrix} \cos \alpha_i & \cos \delta_i \\ \sin \alpha_i & \cos \delta_i \\ \sin \delta_i \end{pmatrix} \quad i = 1, 2, 3, 4, 5 \quad (2-18) \]

\[ \hat{c} = \frac{\hat{c}}{|\hat{c}|} \quad (2-19) \]

where

\[ \hat{c} = \frac{1}{5} \sum_{i=1}^{5} \hat{R}_i \quad (2-20) \]

\[ \hat{c}_B = (R_{em} + h_1) \hat{c} \quad (2-27) \]
\[ \hat{\rho}_B = \hat{R}_{sc} - \hat{C}_B \]  

\[ h_2 - h_1 \geq \frac{\hat{\rho}_B \cdot \hat{C}_B}{|\hat{C}_B|} \geq 0 \]  

\[ \hat{N}_i = \frac{\hat{C}_B \times (\hat{R}_i - \hat{R}_{i+1})}{|\hat{C}_B \times (\hat{R}_i - \hat{R}_{i+1})|} \quad i = 1, 2, 3, 4 \quad (2-26a) \]

\[ \hat{N}_5 = \frac{\hat{C}_B \times (\hat{R}_5 - \hat{R}_1)}{|\hat{C}_B \times (\hat{R}_5 - \hat{R}_1)|} \quad (2-26b) \]

\[ d_i = \hat{N}_i \cdot (\hat{R}_i - \hat{C}_B) \quad i = 1, 2, 3, 4, 5 \quad (2-15) \]

\[ \hat{N}_i \cdot \hat{\rho}_B \leq d_i \quad i = 1, 2, 3, 4, 5 \quad (2-16) \]
3.0 SEMIANALYTICAL ALGORITHM TO PREDICT AOS AND LOS TIMES

This section presents a semianalytic technique for predicting AOS and LOS times. This technique is designed to bound the actual visibility periods and uses a simplified geometry model in conjunction with unperturbed orbital motion. Its principal purpose is to reduce the burden on the precise numerical search by eliminating regions of the S/C orbit where AOS and LOS times are physically impossible.

The basic procedure consists of the following:

a. Determining the parameters of an outer boundary cone which circumscribes the entire area target or space volume.

b. Determining whether the orientation of the S/C orbit plane will permit any intersections of the outer boundary cone with the S/C orbit plane.

c. Assuming intersections are possible for the following:

   (1) Celestial-fixed targets
   (a) Compute the conic intersection points of the outer boundary cone with the S/C orbit plane.
   (b) Compute predictions of AOS and LOS times (\( t_{AOS_k} \) and \( t_{LOS_k} \), respectively) for each revolution based upon these intersection points.

   (2) Earth-fixed targets
   (a) Compute the time of closest approach (TCA) of the S/C position vector with the centerline of the outer boundary cone and the corresponding closest approach angle (\( \gamma_{CA} \)). These computations are performed on a revolution-by-revolution basis.
   (b) If \( \gamma_{CA} \) is less than or equal to the half-cone angle of the outer boundary cone, then compute predictions of AOS and LOS times based upon the times when the S/C lies on the perimeter of the
outer boundary cone. These times are denoted by
\( t_{AOS_K} \) and \( t_{LOS_K} \).

After the AOS and LOS time predictions are determined, the precise numerical search (using the detailed area target and space volume geometry model discussed in sec. 2) can then be performed over the reduced time regions from \( t_{AOS_K} \) to \( t_{LOS_K} \).

The following parameters are required to predict the AOS and LOS times:

- \( t_{\text{start}} \) - start time for the search
- \( t_{\text{end}} \) - end time for the search
- \( \Delta t \) - time increment for the search. For the purposes of this algorithm, \( \Delta t \) will be used as a tolerance for converging upon the AOS and LOS times for Earth-referenced targets.
- \( \vec{R}_{SC_0} \) - S/C position vector in the M50 system at \( t_{\text{start}} \).
- \( \vec{V}_{SC_0} \) - S/C velocity vector in the M50 system at \( t_{\text{start}} \).
- \( [\text{RNP}]_{TEI/M50} \) - RNP matrix to transform from the M50 to the TEI system.
- \( t_e \) - epoch time corresponding to the RNP matrix.

The following subsections provide the necessary equations. Section 3.1 discusses the computation of the outer boundary cone parameters. Section 3.2 discusses the technique to determine whether any AOS or LOS times are possible. Section 3.3 presents the equations to predict AOS and LOS times. Section 3.4 discusses the assumptions and limitations implicit in these equations.

Two reference coordinate systems are used in these computations. The M50 coordinate system is used when dealing with celestial-fixed targets and the TEI system is used for Earth-referenced
targets. The TEI system was selected for Earth-referenced targets in order to model the effects of polar nutation and precession on the Earth's spin axis.

For celestial-fixed targets, the M50 Keplerian elements \( (a, e, i, \Omega, \omega, \text{ and } M_0) \) at the start time will be computed from \( \hat{\mathbf{r}}_{\text{sc}} \) and \( \hat{\mathbf{v}}_{\text{sc}} \).

For Earth-referenced targets, the TEI Keplerian elements at the start time will be computed from \( \hat{\mathbf{r}}_{\text{sc}} \) and \( \hat{\mathbf{v}}_{\text{sc}} \). Where

\[
\hat{\mathbf{r}}_{\text{TEI}} = [\text{RNP}]_{\text{TEI}} \mathbf{r}_{\text{sc}} \tag{3-1}
\]

\[
\hat{\mathbf{v}}_{\text{TEI}} = [\text{RNP}]_{\text{M50}} \mathbf{v}_{\text{sc}} \tag{3-2}
\]
3.1 COMPUTING OUTER BOUNDARY CONE PARAMETERS

The outer boundary cone is defined by

- $\alpha_o$ - right ascension of the centerline
- $\phi_o$ - latitude of the centerline
- $\gamma_A$ - half-cone angle

For Earth-referenced targets these parameters will be defined in the TEI system at $t_{\text{start}}$. For celestial-fixed targets, these parameters will be defined in the M50 system and remain invariant with respect to time.

3.1.1 Earth-Referenced Circles

The unit vector along the centerline of the cone is given in the TEI system by

$$\hat{C}_{\text{TEI}} = \begin{bmatrix} \cos \Delta \lambda - \sin \Delta \lambda & 0 \\ \sin \Delta \lambda & \cos \Delta \lambda \\ 0 & 0 & 1 \end{bmatrix}_{\text{TEI}} \begin{bmatrix} \hat{C}_B^G \\ \hat{C}_B \end{bmatrix}$$

(3-3)

where

- $\hat{C}_B^G$ is given by equation 2-1
- $\Delta \lambda$ is given by equation 2-5 with $t$ equal to $t_{\text{start}}$

Thus, the right ascension and latitude of the centerline is given in the TEI system (at the start of the search) by

$$\alpha_o = \tan^{-1} \left( \begin{bmatrix} \frac{C_{\text{TEI}}^x}{C_{\text{TEI}}^y} \\ \frac{C_{\text{TEI}}^y}{C_{\text{TEI}}^x} \end{bmatrix} \right)$$

(3-4)

\[1\] The TEI system is selected for Earth-referenced targets in order to model the polar nutation and precession from M50 reference to the epoch time of the RNP matrix.
\[ \phi_0 = \sin^{-1} \left\{ \hat{C}_{TEI} \right\} \]  \hspace{1cm} (3-5)

where

\[ \hat{C}_{TEI} \Rightarrow \left( \hat{C}_{TEI}^x, \hat{C}_{TEI}^y, \hat{C}_{TEI}^z \right) \]

The half-cone angle, \( \gamma_A \), is given by equation 2-6.

### 3.1.2 Celestial Circles

The right ascension, latitude, and half-cone angle in the M50 system are given via input

\[ \alpha_0 = \alpha_A \] \hspace{1cm} (3-6)

\[ \phi_0 = \delta_A \] \hspace{1cm} (3-7)

### 3.1.3 Earth-Referenced Polygons

The unit vector along the centerline of the outer boundary cone is given in the TEI system by equation 3-3 where equation 2-13 is used to compute \( \hat{C}_B^G \). Equations 3-4 and 3-5 are then used to compute \( \alpha_0 \) and \( \phi_0 \), respectively. The outer boundary cone is centered along the centroid vector and has an angular radius which contains the vertex furthest from the centroid vector (fig. 3-1). Thus, the half-cone angle is given by

\[ \gamma_A = \max \left[ \cos^{-1} \left( \frac{\vec{R}_i^G \cdot \hat{C}_B^G}{\left| \vec{R}_i^G \right| \left| \hat{C}_B^G \right|} \right) \right] \hspace{1cm} i = 1, 2, 3, \ldots n \] \hspace{1cm} (3-8)

where

\[ \max \text{ implies the maximum algebraic value} \]

\[ \vec{R}_i^G \] \hspace{1cm} (3-5)
Figure 3-1.- Circumscribing polygons.
3.1.4 Celestial Polygons

The unit vector along the centerline of the celestial polygon is given by equation 2-19. Equations 3-4 and 3-5 (with \( \hat{\mathbf{C}}_{\text{TEI}} \) replaced by \( \hat{\mathbf{C}} \)) are then used to compute \( \alpha_0 \) and \( \phi_0 \), respectively. The half-cone angle which contains the vertex furthest from the centroid is given by

\[
\gamma_A = \max \left[ \cos^{-1} \left( \hat{\mathbf{R}}_i \cdot \hat{\mathbf{C}} \right) \right] \quad i = 1, 2, 3, \ldots (3-9)
\]

where

\( \hat{\mathbf{R}}_i \) is given by equation 2-18.

3.1.5 Earth-Referenced Space Volumes

The unit vector along the centerline of the outer boundary cone for Earth-referenced space volumes is given by equation 3-3 with \( \hat{\mathbf{C}}_G \) replaced by \( \hat{\mathbf{C}}_G \) (eq. 2-22). Equations 3-4 and 3-5 are then used to compute \( \alpha_0 \) and \( \phi_0 \), respectively. Since the Earth-referenced space volume represents a constant area polyhedron (sec. 2.5), the outer boundary cone to be used for predicting AOS and LOS times circumscribes the space volume planar area at the S/C altitude. However, for noncircular orbits, the S/C altitude varies as a function of time and thus the size of the outer boundary cone would also vary as a function of time. The size of the cone is largest at perigee and smallest at apogee. Since the purpose of the outer boundary cone is to bound the actual visibility region, the half-cone angle at the S/C perigee altitude will be computed and held constant with respect to time. The vector from the center of the Earth to each vertex of the space volume projected to the S/C perigee altitude, \( \hat{\mathbf{R}}_i^G \), (fig. 3-2) is given by

\[
\hat{\mathbf{R}}_i^G = \hat{\mathbf{R}}_i^G + \left[ \mathbf{R}_P - \left| \hat{\mathbf{C}}_B \right| \right] \frac{\hat{\mathbf{C}}_G}{\left| \hat{\mathbf{C}}_B \right|} \quad i = 1, 2, 3, \ldots n (3-10)
\]
Space volume boundary at perigee altitude

Vertex furthest from centroid

Cone circumscribing space volume at perigee altitude

Lower boundary centroid

Space volume lower boundary

Figure 3-2.- Circumscribing space volumes.
where

\[ \hat{R}_i^G \text{ is given by equation } 2-12 \]

\[ \hat{c}_B^G \text{ is given by equation } 2-23 \]

\[ R_p \text{ is the S/C perigee radius} \]

\[ = a(1 - e) \quad (3-11) \]

Thus the half-cone angle which contains the vertex furthest from the centerline of the cone is given by

\[ \gamma_A = \max \left[ \cos^{-1}\left( \frac{\hat{R}_i^G \cdot \hat{c}_B^G}{|\hat{R}_i^G| |\hat{c}_B^G|} \right) \right] i = 1, 2, 3, \ldots n \quad (3-12) \]

3.1.6 Celestial-Fixed Space Volumes

The unit vector along the centerline of the outer boundary cone for celestial-fixed space volumes is given by equation 2-19.

Equations 3-4 and 3-5 (with \( \hat{c}_{TEI} \) replaced by \( \hat{c} \)) are then used to compute \( \alpha_0 \) and \( \phi_0 \) respectively. Since the celestial-fixed space volume also represents a constant area polyhedron (sec. 2.6), the outer boundary cone to be used for predicting AOS and LOS times circumscribes the space volume planar area at the S/C perigee altitude (sec. 3.1.5). The vector from the center of the Earth to each vertex of the space volume at the S/C perigee altitude, \( \hat{R}_i^' \), is given by

\[ \hat{R}_i^' = \hat{R}_i + \left[ R_p - |\hat{c}_B^G| \right] \hat{c} \quad (3-13) \]
where

\( \hat{\mathbf{R}}_i \) is given by equation 2-18
\( \mathbf{\hat{C}}_B \) is given by equation 2-27
\( R_p \) is the S/C perigee radius
(equation 3-11)

Thus

\[
\gamma_A = \max \left[ \cos^{-1} \left( \frac{\hat{\mathbf{R}}_i \cdot \mathbf{\hat{C}}}{|\hat{\mathbf{R}}_i|} \right) \ i = 1, 2, 3, \ldots n \right] \quad (3-14)
\]
3.2 DETERMINING WHETHER AOS AND LOS TIMES ARE POSSIBLE

The following computations will determine whether the relative orientation of the S/C orbit plane and the outer boundary cone will permit any intersections between these two figures. If no intersections are possible, then no AOS or LOS times are possible. Since the Earth-referenced targets rotate with the Earth and the celestial-fixed targets remain inertially fixed, two different types of tests are required. The celestial-fixed situation is the simplest and is discussed first.

3.2.1 Celestial-Fixed Targets

Figure 3-3 illustrates a celestial-fixed outer boundary cone and a S/C orbit plane. The S/C orbit plane is perpendicular to the unit angular momentum vector which is given by

\[ \hat{h} = \frac{\hat{R}_{SC_o} \times \hat{V}_{SC_o}}{|\hat{R}_{SC_o} \times \hat{V}_{SC_o}|} \quad (3-15) \]

where

\[ \hat{R}_{SC_o} \quad \text{and} \quad \hat{V}_{SC_o} \]

are the S/C position and velocity vectors, respectively, in the M50 system at the beginning of the search.

The centerline of the outer boundary cone, \( \hat{C} \), is given in the M50 system by

\[ \hat{C} = \begin{pmatrix} \cos \alpha_o & \cos \phi_o \\ \sin \alpha_o & \cos \phi_o \\ \sin \phi_o \end{pmatrix} \quad (3-16) \]
Figure 3-3.- Intersection of celestial-fixed outer boundary cone with S/C orbit plane.
where
\( \phi_o \) and \( \phi_o \) were computed using the procedure discussed in
section 3.1.

The intersection points of the outer boundary cone with the S/C
orbit plane are denoted by \( \hat{I}_1 \) and \( \hat{I}_2 \) on figure 3-3. These
vectors can be found by solving the following conic intersection
equations\(^1\)

\[
\begin{align*}
\hat{h} \cdot \hat{I} &= 0 \\
\hat{C} \cdot \hat{I} &= \cos \gamma_A \\
\hat{I} \cdot \hat{I} &= 1
\end{align*}
\]

(3-17a)  
(3-17b)  
(3-17c)

These equations provide 0, 1, or 2 solutions for \( \hat{I} \). If no
solutions are produced, then the relative orientation of the
outer boundary cone and the S/C orbit plane does not permit
intersections, and no AOS or LOS times are possible. If only
one solution is produced, the AOS and LOS times are the same.
If two solutions are produced, the AOS and LOS times predictions
are made using the method outlined in section 3.3.

\(^1\)The method for solving these equations is presented in
appendix B.
3.2.2 Earth-Referenced Targets

Figure 3-4 illustrates an Earth-referenced outer boundary cone at a particular instant of time. This cone rotates about the Earth's polar axis. Hence, over one sidereal day, the volume of space swept out by the outer boundary cone describes a spherical sector whose upper and lower limits are determined by $\gamma_A$ and $\phi_0$ (fig. 3-5). Intersections of the S/C orbit plane with this volume of space are possible only if the S/C orbit inclination exceeds the lower limit of the spherical sector.

For posigrade and polar orbits, the S/C orbit plane will intersect the spherical sector only if

$$i \geq |\phi_0| - \gamma_A$$  

(3-18a)

For retrograde orbits, intersections are possible only if

$$180 - i \geq |\phi_0| - \gamma_A$$  

(3-18b)

where

$$i$$ is the S/C orbit inclination in the TEI system.

If equation 3-18 is not satisfied, then no intersections are possible and thus no AOS or LOS times are possible. If equation 3-18 is satisfied, then the approach presented in section 3.3 is used to predict AOS and LOS times.

---

1 The absolute value sign is used in these equations to accommodate targets in both the northern and southern hemispheres.
Figure 3-4. - Earth-referenced outer boundary cone.
Figure 3-5.- Spherical sector swept out by Earth-referenced outer boundary cone.
3.3 PREDICTING AOS AND LOS TIMES

The following subsections present the equations necessary to predict AOS and LOS times for celestial-fixed and Earth-referenced targets. These equations assume that the tests performed in section 3.2 have indicated that intersections of the outer boundary cone with the S/C orbit plane are possible.

3.3.1 Celestial-Fixed Targets

The intersection vectors, \( \hat{I}_1 \) and \( \hat{I}_2 \) (sec. 3.2.1), can be used directly to predict the AOS and LOS times. Figure 3-6 illustrates the geometry. The unit vector in the direction of the ascending node, \( \hat{\Omega} \), is given by

\[
\hat{\Omega} = \begin{bmatrix}
\cos \Omega \\
\sin \Omega \\
0
\end{bmatrix}
\]  

(3-19)

where

\[
\Omega = \text{right ascension of the ascending node in the M50 system}
\]

The angles between \( \hat{\Omega} \) and the two intersection vectors are given by

\[
u_1 = \tan^{-1} \left( \frac{\hat{h} \cdot (\hat{\Omega} \times \hat{I}_1)}{\hat{\Omega} \cdot \hat{I}_1} \right)
\]

(3-20a)

\[
u_2 = \tan^{-1} \left( \frac{\hat{h} \cdot (\hat{\Omega} \times \hat{I}_2)}{\hat{\Omega} \cdot \hat{I}_2} \right)
\]

(3-20b)
Figure 3-6.- Definition of intersection points.
One of these angles corresponds to the AOS point and the other to the LOS point. The appropriate angle can be determined by noting that the visibility period must be less than or equal to one half of the orbit.

Thus let

$$K = \tan^{-1}\left(\frac{\hat{h} \cdot (\hat{I}_1 \times \hat{I}_2)}{\hat{I}_1 \cdot \hat{I}_2}\right)$$  \hspace{1cm} (3-21)

If $K > 0$

$$u_{AOS} = u_1$$  \hspace{1cm} (3-22a)

$$u_{LOS} = u_2$$  \hspace{1cm} (3-23a)

If $K < 0$

$$u_{AOS} = u_2$$  \hspace{1cm} (3-22b)

$$u_{LOS} = u_1$$  \hspace{1cm} (3-23b)

The true anomalies of the AOS and LOS points are given by

$$f_{AOS/LOS} = u_{AOS/LOS} - \omega$$  \hspace{1cm} (3-24)

where

$\omega$ is the argument of perigee of the S/C orbit.

These can be converted to AOS and LOS times by

$$t_{AOS/LOS} = \frac{M_{AOS/LOS}}{M} + \tau$$  \hspace{1cm} (3-25)

where

$$M_{AOS/LOS} = E_{AOS/LOS} - e \sin E_{AOS/LOS}$$  \hspace{1cm} (3-26)
\[ E_{\text{AOS/LOS}} = 2 \tan^{-1} \left( \frac{1 - e}{1 + e} \tan \left( \frac{\varepsilon_{\text{AOS/LOS}}}{2} \right) \right) \]  

(3-27)

\[ M = \frac{360}{T} \quad \text{(in degrees per unit of time)} \]  

(3-28)

\[ T = 2 \pi \sqrt{\frac{a^3}{\mu}} \]  

(3-29)

\[ t = t_{\text{start}} + \frac{M_0}{M} + (R - 1) T \]  

(3-30)

\( R \) is the revolution number measured with respect to \( t_{\text{start}} \)

\( a \) is the semimajor axis of the S/C orbit

\( e \) is the eccentricity of the S/C orbit

\( M_0 \) is the S/C mean anomaly at \( t_{\text{start}} \)

\( \mu \) is the gravitational constant

Since both the orbit plane and the celestial-fixed target remain inertially fixed, \( u_{\text{AOS}} \) and \( u_{\text{LOS}} \) remain constant. Thus, the AOS and LOS times for future revolutions differ by only the orbital period. The AOS and LOS times for the subsequent \( N \) revolutions are given by

\[ t_{\text{AOS/LOS}_{K+1}} = t_{\text{AOS/LOS}_K} + T \quad K = 1, 2, 3 \ldots N \]  

(3-31)
3.3.2 Earth-Referenced Targets

Computation of AOS and LOS times for Earth-referenced targets is somewhat more complicated than for celestial-fixed targets since the orientation of the Earth-referenced target varies as a function of time. The following procedure will be used to estimate the AOS and LOS times:

a. The TCA of the center of the Earth-referenced outer boundary cone with the S/C position vector will be determined along with the corresponding closest approach angle, $\gamma_{CA}$. These computations will be performed on a revolution-by-revolution basis.

b. The closest approach angle from the previous step will be compared with $\gamma_A$. If $\gamma_{CA} > \gamma_A$, then no AOS or LOS times are possible for that revolution, and the previous step will be repeated for the next revolution. If $\gamma_{CA} \leq \gamma_A$, then AOS and LOS times will be computed by solving for the times when the angle between the center of the outer boundary cone and the S/C position vector is equal to $\gamma_A$.

3.3.2.1 Determining TCA

The closest approach point occurs when the S/C position vector and the center of the outer boundary cone lie in the same hemisphere and are coplanar (fig. 3-7). Both the S/C and the center of the outer boundary cone are moving with time. The center of the cone is rotating about the Earth's Z axis at the Earth's rotation rate. The S/C is moving about the orbit angular momentum vector at a rate which is dependent upon both the orbit period and eccentricity. As a result, the TCA cannot be analytically determined for noncircular/nonequatorial orbits. The TCA can, however, be determined iteratively by the following:
Figure 3-7.- Closest approach point.
a. Estimating the closest approach point (for the first revolution, this estimate would be based upon \( t_{\text{start}} \); for subsequent revolutions, this estimate would be based upon the TCA of the previous revolution plus one orbital period).

b. Determining the time at which the S/C will reach this point.

c. Updating the closest approach point based upon the time estimate from the previous step.

d. Repeating the two previous steps until the difference in time between two successive iterations is less than or equal to the time step, \( \Delta t \).

The following discussion develops the equations necessary to compute the TCA.

The unit vector from the center of the Earth to the center of the outer boundary cone is given by

\[
\hat{C} = \begin{bmatrix}
\cos \alpha \cos \phi_o \\
\sin \alpha \cos \phi_o \\
\sin \phi_o
\end{bmatrix}
\]  

(3-32)

where

\[
\alpha = \alpha_o + \omega_e (t - t_{\text{start}})
\]

(3-33)

\( \alpha_o \) is the initial right ascension of the centerline (sec. 3.1)

\( \omega_e \) is the Earth rotation rate

\( \phi_o \) is the latitude of the centerline (sec. 3.1)
The closest approach point, \( \hat{r}_{CA'} \), occurs when

\[
\hat{r}_{CA} = \frac{\hat{h} \times \hat{C}}{|\hat{h} \times \hat{C}|} \times \hat{h}
\]

(3-34)

where \( \hat{h} \) is the angular momentum vector in the TEI system (computed via eq. 3-15 using \( \hat{R}^{TEI}_{sco} \) and \( \hat{v}^{TEI}_{sco} \) from eqs. 3-1 and 3-2).

The angle from the ascending node to this point, \( u_{CA} \), is given by

\[
u_{CA} = \tan^{-1} \left( \frac{\hat{h} \cdot (\hat{\Omega} \times \hat{r}_{CA})}{\hat{\Omega} \cdot \hat{r}_{CA}} \right)
\]

which can be simplified to

\[
\nu_{CA} = \tan^{-1} \left( \frac{\hat{h} \cdot (\hat{\Omega} \times \hat{C})}{\hat{\Omega} \cdot \hat{C}} \right)
\]

(3-35)

where \( \hat{\Omega} \) is computed by equation 3-19 using the right ascension of the ascending node in the TEI system.

The TCA, \( \tau_{CA} \), is then given by

\[
f_{CA} = \nu_{CA} - \omega
\]

(3-36)

\[
E_{CA} = 2 \tan^{-1} \left( \sqrt{\frac{1-e'}{1+e'}} \tan \left( \frac{f_{CA}}{2} \right) \right)
\]

(3-37)

\[
M_{CA} = E_{CA} - e \sin E_{CA}
\]

(3-38)

\[
\tau_{CA} = \frac{M_{CA}}{M} + \tau
\]

(3-39)
where

\[ \tau \] is given by equation 3-30
\[ M \] is given by equation 3-28
\[ e \] is the orbit eccentricity
\[ \omega \] is the argument of perigee

This time would then be used in equations 3-32 and 3-33 to update the value of \( \hat{C} \). Equations 3-35 through 3-39 would be repeated until the value of \( t_{CA} \) between two successive iterations is less than or equal to the time step, \( \Delta t \). After the TCA has been found, the corresponding closest approach angle, \( \gamma_{CA} \), would be computed.

The unit vector from the center of the Earth to the S/C is given by

\[
\hat{r} = \begin{pmatrix}
\cos \Omega \cos \varphi - \sin \Omega \sin \varphi \cos \iota \\
\sin \Omega \cos \varphi + \cos \Omega \sin \varphi \cos \iota \\
\sin \varphi \sin \iota
\end{pmatrix}
\]

(3-40)

The closest approach angle is given by

\[
\gamma_{CA} = \cos^{-1} \left( \frac{\hat{r}_{CA} \cdot \hat{C}_{CA}}{\|\hat{r}_{CA}\| \|\hat{C}_{CA}\|} \right)
\]

(3-41)

where

\( \hat{r}_{CA} \) is given by equation 3-40 evaluated at \( \varphi_{CA} \)
\( \hat{C}_{CA} \) is given by equations 3-32 and 3-33 evaluated at \( t_{CA} \).
If $\gamma_{CA}$ is greater than the outer boundary cone angle, $\gamma_A$ (sec. 3.1), then no AOS or LOS times are possible for this revolution. In this event, the procedure outlined is repeated for the next revolution. The initial estimate of TCA for the next revolution would be equal to the TCA for the current revolution plus one orbital period. The process continues until either $\gamma_{CA} \leq \gamma_A$ or the TCA exceeds the end time of the search, $t_{end}$. If $\gamma_{CA} < \gamma_A$, estimates of AOS and LOS times are made using the method outlined in section 3.3.2.2.

3.3.2.2 Determining AOS and LOS Times

After a feasible TCA has been found, the AOS and LOS times can be computed by solving the following equations.

$$\hat{r} \cdot \hat{C} = \cos \gamma_A$$  \hspace{1cm} (3-42)

Substituting equations 3-40 and 3-32 into this expression and simplifying yields

$$\cos (\Omega - \alpha) \cos \phi_o \cos u - \sin (\Omega - \alpha) \cos \phi_o \cos i \sin u$$

$$+ \sin \phi_o \sin i \sin u = \cos \gamma_A$$  \hspace{1cm} (3-43)

Equation 3-43 cannot be explicitly solved for time for noncircular/non-equatorial orbits. However, this equation can be solved numerically using the method of successive substitution. For convenience, equation 3-43 can be rewritten as

$$A_1 \cos u + B_1 \sin u = C_1$$  \hspace{1cm} (3-44)

where

$$A_1 = \cos \phi_o \cos (\Omega - \alpha)$$  \hspace{1cm} (3-45)

$$B_1 = \sin \phi_o \sin i - \cos \phi_o \cos i \sin (\Omega - \alpha)$$  \hspace{1cm} (3-46)

$$C_1 = \cos \gamma_A$$  \hspace{1cm} (3-47)
Assuming $a$ is constant between iterations, equation 3-44 has the following solution

$$u_{AOS/LOS} = \tan^{-1}\left(\frac{B_1}{A_1}\right) + \cos^{-1}\left(\frac{C_1}{\sqrt{A_1^2 + B_1^2}}\right)$$  \hspace{1cm} (3-48)

The minus sign will produce AOS solutions and the plus sign will produce LOS solutions. The procedure is as follows:

a. Solving for AOS times, $t_{AOS}$
   
   (1) Use initial estimate of $t_{AOS} = t_{CA}$.
   (2) Compute $A_1$, $B_1$, and $C_1$ using equations 3-33, 3-45, 3-46, and 3-47.
   (3) Compute $u_{AOS}$ using equation 3-48 with the minus sign.
   (4) Compute revised estimate of AOS time using equations 3-24 through 3-30.
   (5) If the change in time from the previous iteration is less than $\Delta t$, then a solution has been found. Otherwise, repeat steps (2) through (5) with the revised time estimate.

b. Solving for LOS time, $t_{LOS}$

The solution for LOS time is similar to the procedure for AOS time except that

(1) The initial estimate of LOS time in step a(1) is given by

$$t_{LOS} = t_{CA} + (t_{CA} - t_{AOS})$$ \hspace{1cm} (3-49)

(2) $u_{LOS}$ is computed in step a(3) by using equation 3-48 with the plus sign.

(3) The revised estimate of LOS time is produced in step a(4).
The computations discussed in this section are performed for each revolution in which $\gamma_{CA}$ was found to be less than or equal to $\gamma_A$. In the special case where $\gamma_{CA}$ is exactly equal to $\gamma_A$, the AOS and LOS times are identical and equal to the TCA. Hence computation of AOS and LOS times can be bypassed.
3.4 ASSUMPTIONS AND LIMITATIONS

The following assumptions are implicit in the equations presented in sections 3.1 through 3.3.

a. Unperturbed orbital motion over the time period \( t_{\text{end}} \geq t \geq t_{\text{start}} \).

b. The effects of polar nutation and precession from \( t_e \) to \( t_{\text{end}} \) can be neglected.
4.0 FUNCTIONAL OVERVIEW OF AOS AND LOS TIME COMPUTATIONS

This section provides a functional overview of the ATSVP. The overall procedure for determining precise AOS and LOS times is presented and discussed. The basic procedure consists of predicting the AOS and LOS times using the algorithm presented in section 3 and then refining these predictions by performing a sequential time search using the equations presented in section 2.

The AOS times are defined to be the time point at which the S/C first enters the area target or space volume. If the S/C lies within the area target or space volume at the beginning of the user-specified search period \( t_{\text{start}} \), the first AOS time will be set equal to the start time. LOS times are defined to be the last time point prior to the S/C exiting the area target or space volume. If the S/C lies within the area target or space volume at the end of the user-specified search period \( t_{\text{end}} \), then the last LOS time will be set equal to the end time. It should be noted that the time increment used to perform the sequential time search will limit the accuracy and resolution of the AOS and LOS times (e.g., if the time increment is 1 minute, this implies that the AOS and LOS times will be determined to the nearest minute and that visibility periods of less than 1 minute may be skipped).

The area targets and space volumes defined in section 1 fall into two general categories:

a. Earth-referenced which include
   (1) Earth-referenced circles.
   (2) Earth-referenced polygons.
   (3) Earth-referenced space volumes.

b. Celestial-fixed which include
   (1) celestial-fixed circles.
   (2) celestial-fixed polygons.
   (3) celestial-fixed space volumes.
The logic for computing the AOS and LOS times for each of these categories is presented in the following subsections.

4.1 EARTH-REFERENCED AREA TARGETS AND SPACE VOLUMES

Figure 4-1 provides a functional flowchart for the Earth-referenced area targets and space volumes. The required inputs are:

a. Target table (either the ground target, ground polygon, or ground-fixed space volume table).

b. Target ID.

c. Start and end times and time increment (t_{start}, t_{end}, and Δt, respectively).

d. S/C ephemeris and ephemeris ID.

e. RNP matrix and associated epoch time.

A brief description of the process is provided herein. The heading numbers correspond to the numbered blocks on figure 4-1.

1. The geodetic parameters defining the area target or space volume are obtained from the appropriate target table based upon the input target ID. Sections 2.1.2, 2.3.2, and 2.5.2 describe the specific parameters which are required.

2. The RNP matrix and its associated epoch time are obtained.

3. The geodetic coordinates of the area target or space volume are transformed to the rotating geocentric coordinate system (secs. 2.1.2, 2.3.2, and 2.5.2).

4. The number of visibility periods between t_{start} and t_{end} are determined along with predictions of AOS and LOS times. Section 3 describes this process.

5. A test is performed to determine whether any visibility periods were found. If this test is failed, then no AOS/LOS times are possible and processing is terminated.

6. A loop is established which will process each visibility period found in step 4.
Figure 4-1.- Functional flow for Earth-referenced area targets and space volumes.
7. The current time, \( t \), is initialized to the AOS time prediction, \( t_{A0S} \). The initial value of the precise AOS time, \( AOS_1 \), is set to zero. The AOS/LOS counter, \( i \), is initialized to one.

8. A loop is established which will terminate when \( t \) exceeds the LOS time prediction, \( t_{LOS} \).

9. The S/C position vector (at time \( t \) in the M50 system) is obtained from the ephemeris file based upon the ephemeris ID.

10. The S/C position vector is transformed to the rotating geocentric system and computations are performed to determine whether the S/C lies within the area target or space volume (secs. 2.1.2, 2.3.2, 2.5.2, and app. A).

Based upon the results of these tests, the S/C visibility parameter, \( V \), is set

\[ V > 0 \text{ if the S/C lies within the area target or space volume} \]

\[ V < 0 \text{ if the S/C lies exterior to the area target or space volume} \]

11. A test is performed on \( V \).

If \( V > 0 \)

11.1 a further test is performed on \( AOS_1 \) to determine whether the S/C was also visible during one or more of the previous time steps.

If \( AOS_1 > 0 \)

11.1.1 then the S/C was visible during one or more of the previous time steps. Hence, no transi-

---

1 Depending upon the target geometry and the S/C groundtrack, multiple AOS/LOS times could occur during a single visibility period.
tion has occurred. The current time is incremented by $\Delta t$ and the search continues.

If $AOS_i < 0$
11.1.2 then a transition into the area target or space volume has occurred. The current AOS time, $AOS_i$, is set equal to the current time. The current time is incremented by $\Delta t$ and the search continues.

If $V < 0$
11.2 a further test is performed on $AOS_i$ to determine whether the S/C was visible during the previous time step.

If $AOS_i > 0$
11.2.1 then the S/C was visible during the previous time step and a transition out of the area target or space volume has occurred. The current LOS time, $LOS_i$, is set equal to the time of the previous time step ($t - \Delta t$). The AOS/LOS counter, $i$, is incremented by one. The next AOS time is initialized to zero. The current time is incremented by $\Delta t$ and the search continues.

11.2.2 the S/C was not visible during the preceding time step. Thus no transition has occurred. The current time is incremented by $\Delta t$ and the search continues.

12. At the completion of the sequential time search, a test is made to determine whether the S/C was visible at the end time. If the test is true, the last LOS time is set equal to $t_{LOS_k}$.

4.2 CELESTIAL-FIXED AREA TARGETS AND SPACE VOLUMES

Figure 4-2 illustrates the logic flow for the celestial-fixed area targets and space volumes. This logic is similar to the approach presented in section 4.1. The required inputs are
Figure 4-2.- Functional flow for celestial-fixed area targets and space volumes.
a. Target table (either celestial circles, celestial polygons, or celestial-fixed space volumes table).

b. Target ID.

c. Start and end times and time increment \( t_{\text{start}}, t_{\text{end}}, \) and \( \Delta t \), respectively.

d. S/C ephemeris and ephemeris ID.

A brief description of the process is provided herein. The heading numbers correspond to the numbered blocks on figure 4-2.

1. The parameters defining the celestial-fixed area target or space volume are obtained from the appropriate target table based upon the input target ID. Sections 2.2.2, 2.4.2, and 2.6.2 describe the specific parameters which are required.

2. The number of visibility periods between \( t_{\text{start}} \) and \( t_{\text{end}} \) are determined along with predictions of AOS and LOS times. Section 3 describes this process.

3. A test is performed to determine whether any visibility periods were found. If this test is failed, then no AOS/LOS times are possible and processing is terminated.

4. A loop is established which will process each visibility period found in step 2.

5. The current time, \( t \), is initialized to the AOS time prediction, \( t_{\text{AOS}_K} \). The initial value of the precise AOS time, \( t_{\text{AOS}_1} \), is set to zero. The AOS/LOS counter, \( i \), is initialized to one.

6. A loop is established which will terminate when \( t \) exceeds the LOS time prediction, \( t_{\text{LOS}_K} \).

7. The S/C position vector (at time \( t \) in the M50 system) is obtained from the ephemeris file based upon ephemeris ID.

8. Computations are performed to determine whether the S/C lies within the area target or space volume (secs. 2.2.2, 2.4.2,

---

1 Depending upon the target geometry and the S/C groundtrack, multiple AOS/LOS times could occur during a single visibility period.
2.6.2, and app. A). Based upon the results of these tests, the S/C visibility parameter, $V$, is set

\[ V > 0 \text{ if the S/C lies within the area target or space volume} \]

\[ V < 0 \text{ if the S/C lies exterior to the area target or space volume} \]

9. A test is performed on $V$.

If $V > 0$

9.1 a further test is performed on $AOS_i$ to determine whether the S/C was also visible during one or more of the previous time steps.

If $AOS_i > 0$

9.1.1 then the S/C was visible during one or more of the previous time steps. Hence, no transition has occurred. The current time is incremented by $\Delta t$ and the search continues.

If $AOS_i \leq 0$

9.1.2 then a transition into the area target or space volume has occurred. The current AOS time, $AOS_i$, is set equal to the current time. The current time is incremented by $\Delta t$ and the search continues.

If $V < 0$

9.2 a further test is performed on $AOS_i$ to determine whether the S/C was visible during the previous step.

If $AOS_i > 0$

9.2.1 then the S/C was visible during the previous time step and a transition out of the area target or space volume has occurred. The current LOS time, $LOS_i$, is set equal to the time of the previous time step ($t - \Delta t$). The AOS/LOS counter, $i$, is
incremented by one. The next AOS time is initialized to zero. The current time is incremented by $\Delta t$ and the search continues.

9.2.2 the S/C was not visible during the preceding time step. Thus, no transition has occurred. The current time is incremented by $\Delta t$ and the search continues.

10. At the completion of the sequential time search, a test is made to determine whether the S/C was visible at the end time. If the test is true, the last LOS time is set equal to $t_{LOS_K}$. 
5.0 DETAILED LOGIC FLOW

Figure 5-1 presents the detailed logic flow for the area targets and space volumes processor. This flowchart integrates the functional overview presented in section 4 with the equations and approach presented in sections 2 and 3 and the appendixes.

Annotations are provided on the flowchart to describe the computations and tests that are performed. In order to simplify figure 5-1, several notations have been used to represent corresponding FORTRAN functions, e.g.,

\[
\phi(I) \quad \text{represents an array of longitudes indexed by the variable } I.
\]

\[\hat{R}(I) \quad \text{represents a two-dimensional array [i.e., } R(3,5)] \text{ in which the first dimension represents the Cartesian components of the vector and the second dimension is indexed by the variable } I. \text{ Hence, operations involving vector quantities imply a "DO loop" on the innermost dimension.}\]

\[\hat{N}(J,I) \quad \text{represents a three-dimensional array [i.e., } N(3,5,3)] \text{ in which the first dimension represents the Cartesian components of the vector and the second and third dimensions are indexed by variables } J \text{ and } I, \text{ respectively. Operations involving these vector quantities also imply a DO loop on the innermost dimension.}\]

\[
\sum_{L=1}^{K-1} L \quad \text{implies summation over the indicated range using the variable specified by the lower bound (e.g., the variable } L \text{ will be used to perform the summation over the range } L = 1 \text{ to } L = K - 1).\]
Initialize
NSIDES = 1
NSEG = 1
h(I) = 0
(I = 1, 2, 3, 4, 5)

Obtain search control parameters
t' \text{start}', t' \text{end}', \Delta t
Target ID
Ephemeris ID

Obtain target parameters
a. Earth-referenced circles
\phi(I), \lambda(I), h(I), r_c
b. Celestial circles
\alpha(I), \delta(I), \gamma_A
c. Earth-referenced polygon
NSIDES, NSEG, VO(J)
(\phi(I), \lambda(I), h(I))
(I = 1, 2, ..., NSIDES)
d. Celestial polygon
NSIDES, NSEG, VO(J)
(\alpha(I), \delta(I))
(I = 1, 2, ..., NSIDES)
e. Earth-referenced space volume
NSIDES, NSEG, VO(J)
(\phi(I), \lambda(I), h_1, h_2)
(I = 1, 2, ..., NSIDES)
f. Celestial space volume
NSIDES, NSEG, VO(J)
(\alpha(I), \delta(I), h_1, h_2)
(I = 1, 2, ..., NSIDES)

If Target is Earth-referenced

then 1
Compute vectors to each vertex

else 2

Figure 5-1. - Detailed flowchart.
Earth-referenced targets

1. Obtain RNP matrix and epoch time

\[ [\text{RNP}]_{\text{TEI}}^{\text{M50}} \]

\[ t_e \]

2. RNP matrix table

3. Compute geocentric vectors to each vertex

4. If Target is a circle

5. DO

6. \[ I = 1, \text{NSIDES} \]

7. Compute angular radius (in radians)

\[ \gamma_A = \frac{r_c}{|\hat{\mathbf{R}}(I)|} \]

\[ a_F = \frac{R_{\text{em}}}{\sqrt{\cos^2[\phi(I)] + (1 - F)^2 \sin^2[\phi(I)]}} \]

\[ \hat{\mathbf{R}}(I) = \begin{bmatrix} [h(I) + a_F] \cos [\lambda(I)] \cos [\phi(I)] \\ [h(I) + a_F] \sin [\lambda(I)] \cos [\phi(I)] \\ [h(I) + (1 - F)^2 a_F] \sin [\phi(I)] \end{bmatrix} \]

Figure 5-1.- Continued.
Celestial-fixed targets

**Figure 5-1.** - Continued.
Obtain S/C position ($\mathbf{\hat{R}}_{S/C}$) and velocity ($\mathbf{\hat{V}}_{S/C}$) vectors at ephemeris start.

If Target is Earth-referenced:

Transform $\mathbf{\hat{R}}_{S/C}$, $\mathbf{\hat{V}}_{S/C}$ to Keplerian elements ($a, e, i; \Omega, \omega, M_0$).

Test Target ID:

Compute unit centroid vector:

$$\mathbf{\hat{C}_{TB}} = \frac{\mathbf{C}_{TB}}{||\mathbf{C}_{TB}||}$$

Figure 5-1.—Continued.
Earth-referenced or celestial circles

Earth-referenced or celestial polygon

Earth-referenced space volume

Celestial space volume

Figure 5-1. - Continued.

5-6
TAI Transform unit Earth-referenced centroid vector to TEI system

\[ \alpha_o = \tan^{-1}\left(\frac{\hat{C}_{BY}}{\hat{C}_{BX}}\right) \]

\[ \phi_o = \sin^{-1}\left(\frac{\hat{C}_{BZ}}{\hat{C}_{BY}}\right) \]

Compute right ascension and latitude of centerline

Test Target ID

Compute \( T_a \)

IF Target is celestial-fixed

Determine whether AOS/LOS times are possible

Outer boundary cone does not intersect S/C orbit plane. No AOS/LOS times are possible.

IF END = YES

EXIT

\[ T = 2\pi\sqrt{\frac{a^3}{\mu}} \]

\[ N = \frac{2\pi}{T} \]

\[ \tau = t_{\text{start}} - \frac{M_0}{M} \]

Compute orbital period, orbital rate and time of perigee passage

Figure 5-1.- Continued.
\[ \Delta \lambda = \omega_e (t_{\text{start}} - t_e) \]

\[ \hat{C}_T^B = \begin{bmatrix} \cos (\Delta \lambda) - \sin (\Delta \lambda) & 0 \\ \sin (\Delta \lambda) \cos (\Delta \lambda) & 0 \\ 0 & 0 & 1 \end{bmatrix} \]

Transform unit centroid vector to TEI system

\[ \gamma = \max_{A} \left\{ \frac{\hat{R}(I) \cdot \hat{c}_T^B}{||\hat{R}(I)|| ||\hat{c}_T^B||} \right\} \]

Earth-referenced or celestial polygon

\[ (I = 1, 2, \ldots \text{NSIDES}) \]

\[ \gamma_A = \max \left\{ \frac{\hat{R}(I) \cdot \hat{c}_T^B}{||\hat{R}(I)|| ||\hat{c}_T^B||} \right\} \]

\[ (I = 1, 2, \ldots \text{NSIDES}) \]

Earth-referenced or celestial space volume

\[ \hat{R}(I) = \hat{R}(I) + \left[ R_p - ||\hat{c}_T^B|| \right] \frac{\hat{c}_T^B}{||\hat{c}_T^B||} \]

Earth-referenced or celestial space volume

\[ \gamma_A = \max \left\{ \frac{\hat{R}(I) \cdot \hat{c}_T^B}{||\hat{R}(I)|| ||\hat{c}_T^B||} \right\} \]

\[ (I = 1, 2, \ldots \text{NSIDES}) \]

Figure 5-1. Continued.
Figure 5-1.- Continued.
Figure 5-1. Continued.

Retrograde orbit

IF $\pi - i \geq |\phi_o| - \gamma_A$

then

then $\\\text{END = NO}$

else $\\\text{END = YES}$

Outer boundary cone intersects S/C orbit plane at some time

Posigrade or polar orbit

IF $i \geq |\phi_o| - \gamma_A$

then $\\\text{END = NO}$

else $\\\text{END = YES}$

Outer boundary cone does not intersect S/C orbit plane

then $\\\text{END = NO}$

else $\\\text{END = YES}$

Outer boundary cone intersects S/C orbit plane at some time

Outer boundary cone does not intersect S/C orbit plane
IF Target is celestial-fixed

then

11 Compute AOS/LOS time predictions

else

14

IF REV = 0

EXIT

No AOS/LOS times in user-specified search period

then

20 Compute centroid


\[ \mathbf{c}_B = \mathbf{r}(1) \]

Decode vertex ordering integer

else

21

DO

J = 1, NSEG

DO

I = 1, NSEG

Compute lower boundary centroid, unit outward normal vectors, and distance to each side

DO

K = 1, REV

Perform time search for precise AOS/LOS times

IF IAOS > 0

TLOS = t_{end}

Set final LOS time equal to end time

Generate final LOS entry

AOS/LOS table

Test if S/C was visible at end of search.

EXIT

Figure 5-1.- Continued.
\[ S = \frac{1}{2} (\pi/2 + \gamma_A + \gamma) \]

\[ K = \sin(S - \frac{\pi}{2}) \sin(S - \gamma_A) \sin(S - \gamma) \]

\[ B = 2 \tan^{-1} \left( \frac{K}{\sin(s)} \right) \]

\[ \hat{\imath}_2 = \frac{\hat{h} \times \hat{c}}{|\hat{h} \times \hat{c}|} \]

\[ \hat{\imath}_1 = \hat{\imath}_2 \times \hat{h} \]

\[ \hat{I}_1 = \begin{bmatrix} \hat{\imath}_{1x} & \hat{\imath}_{1y} & \hat{\imath}_{1z} \end{bmatrix} \begin{bmatrix} \cos(B) \\ \sin(B) \\ 0 \end{bmatrix} \]

\[ \hat{I}_2 = \begin{bmatrix} \hat{\imath}_{2x} & \hat{\imath}_{2y} & \hat{\imath}_{2z} \end{bmatrix} \begin{bmatrix} \cos(B) \\ 0 \\ -\sin(B) \end{bmatrix} \]

\[ \hat{\Omega} = \begin{bmatrix} \cos(\Omega) \\ \sin(\Omega) \\ 0 \end{bmatrix} \]

\[ u_1 = \tan^{-1} \left( \frac{\hat{h} \cdot (\hat{\Omega} \times \hat{I}_1)}{\hat{\Omega} \cdot \hat{I}_1} \right) \]

\[ u_2 = \tan^{-1} \left( \frac{\hat{h} \cdot (\hat{\Omega} \times \hat{I}_2)}{\hat{\Omega} \cdot \hat{I}_2} \right) \]

\[ \kappa = \tan^{-1} \left( \frac{\hat{h} \cdot (\hat{I}_1 \times \hat{I}_2)}{\hat{I}_1 \cdot \hat{I}_2} \right) \]

Figure 5-1.- Continued.
Determine AOS/LOS points

\[
\begin{align*}
\text{IF} & \quad \kappa \geq 0 \\
\text{then} & \quad u_{\text{AOS}} = u_1 \\
& \quad u_{\text{LOS}} = u_2 \\
\text{else} & \quad u_{\text{AOS}} = u_2 \\
& \quad u_{\text{LOS}} = u_1
\end{align*}
\]

\[
\begin{align*}
\frac{d}{dt} \text{AOS} &= u_{\text{AOS}} - \omega \\
\frac{d}{dt} \text{LOS} &= u_{\text{LOS}} - \omega \\
E_{\text{AOS}} &= 2 \tan^{-1} \left( \frac{1-e}{1+e} \tan \left( \frac{\frac{d}{dt} \text{AOS}}{2} \right) \right) \\
E_{\text{LOS}} &= 2 \tan^{-1} \left( \frac{1-e}{1+e} \tan \left( \frac{\frac{d}{dt} \text{LOS}}{2} \right) \right) \\
M_{\text{AOS}} &= E_{\text{AOS}} - e \sin (E_{\text{AOS}}) \\
M_{\text{LOS}} &= E_{\text{LOS}} - e \sin (E_{\text{LOS}}) \\
t_{\text{AOS}} &= \frac{M_{\text{AOS}}}{M} + \tau \\
t_{\text{LOS}} &= \frac{M_{\text{LOS}}}{M} + \tau
\end{align*}
\]

Compute estimates of initial AOS/LOS times

Initialize
\[\text{REV} = 1\]

Do while \(t_{\text{LOS}}^{(\text{REV})} < t_{\text{end}}\)

Predict AOS/LOS times over time span \(t_{\text{end}} > t > t_{\text{start}}\)

Figure 5-1.- Continued.
\[ t_{AOS}(REV) = t_{AOS_0} + (REV-1) T \]
\[ t_{LOS}(REV) = t_{LOS_0} + (REV-1) T \]

**Compute AOS/LOS time estimates for each revolution**

**IF**
\[ t_{AOS}(REV) < t_{start} \]
**THEN**
\[ t_{AOS}(REV) = t_{start} \]
**ELSE**

**IF**
\[ t_{AOS}(REV) > t_{end} \]
**THEN**
\[ REV = REV - 1 \]
End DO WHILE
**ELSE**

**IF**
\[ t_{LOS}(REV) > t_{end} \]
**THEN**
\[ t_{LOS}(REV) = t_{end} \]
End DO WHILE
**ELSE**

\[ REV = REV + 1 \]

**Increment revolution counter**

**Figure 5-1.- Continued.**
Initialize 
$\text{REV} = 1$
$t_{CA} = t_{\text{start}}$
$\text{MAXITR} = 20$

$\hat{h} = \frac{\hat{r}_{SC_0} \times \hat{v}_{SC_0}}{\left| \hat{r}_{SC_0} \times \hat{v}_{SC_0} \right|}$

Compute unit angular momentum vector and ascending node vector

$\hat{\Omega} = \begin{bmatrix} \cos (\Omega) \\ \sin (\Omega) \\ 0 \end{bmatrix}$

DO WHILE $t_{LOS} (\text{REV}) < t_{\text{end}}$

DO $N = 1, \text{MAXITR}$

Compute TCA

Figure 5-1.- Continued.
\[ \alpha = \alpha_0 + \omega_e (t_{CA} - t_{\text{start}}) \]

\[ \hat{C} = \begin{bmatrix} \cos(\alpha) \cos(\phi_0) \\ \sin(\alpha) \cos(\phi_0) \\ \sin(\phi_0) \end{bmatrix} \]

\[ u_{CA} = \tan^{-1}\left( \frac{h \cdot (\hat{\Omega} \times \hat{C})}{\hat{\Omega} \cdot \hat{C}} \right) \]

\[ f_{CA} = u_{CA} - \omega \]

\[ E_{CA} = 2 \tan^{-1}\left( \frac{1-e}{1+e} \tan\left( \frac{f_{CA}}{2} \right) \right) \]

\[ M_{CA} = E_{CA} - e \sin(E_{CA}) \]

\[ t_{CA}^* = \frac{M_{CA}}{\dot{M}} + \tau + (\text{REV-1}) \tau \]

Compute new estimate of TCA

\[ H \]

TCA iteration has converged

then \[ t_{CA} = t_{CA}^* \]

End DO loop on N

IF \[ |t_{CA}^* - t_{CA}| \leq \Delta t \]

Update TCA time estimate and continue iteration

else \[ t_{CA} = t_{CA}^* \]

Figure 5-1.- Continued.
Current TCA exceeds search end time by T. No further AOS/LOS times are required.

\[
\begin{align*}
\hat{r}_{CA} &= \begin{pmatrix}
\cos(\Omega) \cos(u_{CA}) - \sin(\Omega) \sin(u_{CA}) \cos(i) \\
\sin(\Omega) \cos(u_{CA}) + \cos(\Omega) \sin(u_{CA}) \cos(i) \\
\sin(u_{CA}) \sin(i)
\end{pmatrix} \\
\gamma_{CA} &= \cos^{-1} \left( \hat{r}_{CA} \cdot \hat{C} \right)
\end{align*}
\]

Compute closest approach angle

IF \( \gamma_{CA} \leq \gamma_A \)

Compute AOS/LOS time predictions

Increment \( t_{CA} = t_{CA} + T \)

Figure 5-1. - Continued.
Initialize
\[ t_{\text{AOS}}(\text{REV}) = t_{\text{CA}} \]
\[ C_1 = \cos(Y_A) \]

DO
\[ N = 1, \text{MAXITR} \]

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\[ \alpha = \alpha_0 + \omega_e \left[ t_{\text{AOS}}(\text{REV}) - t_{\text{start}} \right] \]
\[ A_1 = \cos(\phi_o) \cos(\Omega - \alpha) \]
\[ B_1 = \sin(\phi_o) \sin(i) - \cos(\phi_o) \cos(i) \sin(\Omega - \alpha) \]

\[ u_{\text{AOS}} = \tan^{-1} \left( \frac{B_1}{A_1} \right) - \cos^{-1} \left( \frac{C_1}{\sqrt{A_1^2 + B_1^2}} \right) \]

\[ f_{\text{AOS}} = u_{\text{AOS}} - \omega \]
\[ E_{\text{AOS}} = 2 \tan^{-1} \left( \frac{1 - e}{1 + e} \tan \left( \frac{f_{\text{AOS}}}{2} \right) \right) \]

\[ M_{\text{AOS}} = E_{\text{AOS}} - e \sin(E_{\text{AOS}}) \]

\[ t'_{\text{AOS}} = \frac{M_{\text{AOS}}}{\dot{M}} + \tau + (\text{REV}-1) T \]

IF
\[ |t'_{\text{AOS}} - t_{\text{AOS}}(\text{REV})| \leq \Delta t \]

then
\[ t_{\text{AOS}}(\text{REV}) = t'_{\text{AOS}} \]
End DO loop on N

else
\[ t_{\text{AOS}}(\text{REV}) = t_{\text{AOS}} \]

Update AOS time estimate and continue iteration

Figure 5-1.- Continued.
Initialize
\[ t_{LOS}(REV) = t_{CA} + [t_{CA} - t_{AOS}(REV)] \]

If
\[ t_{AOS}(REV) < t_{\text{start}} \]
then
\[ t_{AOS}(REV) = t_{\text{start}} \]

Limit AOS time estimate to start of search

If
\[ t_{AOS}(REV) > t_{\text{end}} \]
then
\[ \text{REV} = \text{REV} - 1 \]
End DO WHILE

Terminate outer DO loop if AOS time exceeds end of search

DO
\[ N = 1, \text{MAXITR} \]
End DO WHILE

Compute LOS

\[ \text{REV} = \text{REV} + 1 \]

Limit LOS time estimate to end of search. Terminate outer DO loop when LOS exceeds end of search

\[ t_{LOS}(REV) = t_{\text{end}} \]
End DO WHILE

Figure 5-1.— Continued.
\[ \alpha = \alpha_0 + \omega_0 [t_{\text{LOS}}(\text{REV}) - t_{\text{start}}] \]

\[ A_1 = \cos (\phi_0) \cos (\Omega - \alpha) \]

\[ B_1 = \sin (\phi_0) \sin (i) - \cos (\phi_0) \]

\[ \cos (i) \sin (\Omega - \alpha) \]

\[ u_{\text{LOS}} = \tan^{-1}\left(\frac{B_1}{A_1}\right) + \tan^{-1}\left(\frac{C_1}{\sqrt{A_1^2 + B_1^2}}\right) \]

\[ f_{\text{LOS}} = u_{\text{LOS}} - \omega \]

\[ E_{\text{LOS}} = 2 \tan^{-1}\left(\frac{1-e^2}{1+e} \tan\left(\frac{f_{\text{LOS}}}{2}\right)\right) \]

\[ M_{\text{LOS}} = E_{\text{LOS}} - e \sin (E_{\text{LOS}}) \]

\[ t_{\text{LOS}} = \frac{M_{\text{LOS}}}{M} + \tau + (\text{REV} - 1) \tau \]

\[ \text{Compute new estimate of LOS time.} \]

\[ \text{Update LOS time estimate and continue iteration} \]

\[ \text{then} \]

\[ t_{\text{LOS}}(\text{REV}) = t_{\text{LOS}} \]

\[ \text{End DO loop on N} \]

\[ \text{LOS iteration has converged} \]

\[ \text{else} \]

\[ t_{\text{LOS}}(\text{REV}) = t_{\text{LOS}} \]

\[ \text{Update LOS time estimate and continue iteration} \]

\[ \text{end} \]

\[ \text{Figure 5-1.- Continued.} \]
DO IV = 1,5
IF IV ≤ 0
END DO
FO loop on N
Loop on N ends normally
NMAX = 5

IVO(1,J) = VO(J)/10(NMAX-1)

DO K=2, NMAX
IVO(K,J) = VO(J) - \sum_{L=1}^{K-1} IVO(L,J) 10(NMAX-L)
\frac{10(NMAX-K)}{10(NMAX-L)}

Save number of sides for this segment

NVO(J) = NMAX

Decode vertex ordering integer and determine number of sides in each segment of the area target or space volume

Loop on N is terminated when highest non zero value of VO(J) is found

Decode vertex ordering integer into vertex numbers for each vertex of each segment

Figure 5-1.- Continued.
Set NV equal to number of sides for this segment

DO
J = 1, NV

DO
K = 1, 3

L = IVO(J, I)
RI(K, J) = R(K, L)

Extract vectors to each vertex for this segment and arrange into array, RI, according to decoded vertex ordering integer

NVMI = NV - 1

DO
J = 1, NVMI

IF Target is area target

22
Compute lower boundary centroid

23
Compute unit outward normal vectors

IF Target is area target

24
Compute last unit outward normal vector

d(J, I) = \( \hat{n}(J, I) \cdot \left[ \hat{RI}(J) - \hat{C}_B(I) \right] \)

Compute perpendicular distance from lower boundary centroid to each side of each segment

Figure 5-1.- Continued.
Figure 5-1.- Continued.
then \( n(J,I) = \frac{\hat{\mathbf{R}}_I(J+1) \times \hat{\mathbf{R}}_I(J)}{|\hat{\mathbf{R}}_I(J+1) \times \hat{\mathbf{R}}_I(J)|} \)

\[ \text{TP} = \hat{\mathbf{R}}_I(J) - \hat{\mathbf{R}}_I(J+1) \]

else

\[ \hat{n}(J,I) = \frac{\hat{c}_B(I) \times \mathbf{TP}}{|\hat{c}_B(I) \times \mathbf{TP}|} \]

then \( \hat{n}(NV,I) = \frac{\hat{\mathbf{R}}_I(1) \times \hat{\mathbf{R}}_I(NV)}{|\hat{\mathbf{R}}_I(1) \times \hat{\mathbf{R}}_I(NV)|} \)

\[ \text{TP} = \hat{\mathbf{R}}_I(NV) - \hat{\mathbf{R}}_I(1) \]

else

\[ \hat{n}(NV,I) = \frac{\hat{c}_B(I) \times \mathbf{TP}}{|\hat{c}_B(I) \times \mathbf{TP}|} \]

Figure 5-1.- Continued.
Set start time and initial visibility parameter to not visible

\( \text{IVIS} = 0 \)

**Initialize**

\( t = t_{AOS}^{(K)} \)

**DO WHILE**

\( t < t_{LOS}^{(K)} \)

Obtain S/C position vector (\( \vec{R}_{sc} \)) at time, \( t \), from S/C ephemeris file based upon ephemeris ID

S/C ephemeris file

**IF**

Target is Earth-referenced

\[ \Delta \lambda = \omega_e (t - t_e) \]

\[ \vec{R}_{sc} = \begin{bmatrix} \cos(\Delta \lambda) & \sin(\Delta \lambda) & 0 \\ -\sin(\Delta \lambda) & \cos(\Delta \lambda) & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} \vec{R}_{sc} \end{bmatrix}_{TEI} \]

Transform S/C position vector from M50 system to geocentric system

Figure 5-1.- Continued.
Figure 5-1. - Continued.
Initialize boundary test parameter

```
ILOW = 0
```

```
DO I = 1, NSEG
  \[ \hat{\rho} = \mathbf{R}_{sc} - \mathbf{c}_B(I) \]
  \[ b = \hat{\rho} \cdot \frac{\mathbf{c}_B(I)}{|\mathbf{c}_B(I)|} \]
```

Test boundary

```
IF b < 0 .OR. b > h_2 - h_1
```

S/C is either below lower boundary or above upper boundary for this segment. Increment test parameter

```
ILOW = ILOW + 1
```

Figure 5-1.- Continued.
DO
I = 1, NSEG

Initialize containment parameter for each segment to not contained.
Set NV to number of sides for this segment.

Compute slant range from lower boundary centroid for each segment.

DO
J = 1, NV

Compute dot product of unit normal vector (to each side of each segment) and slant range vector.

IF
b ≤ d (J, I)

S/C is contained within this side of this segment. Increment containment counter.

IF
IPL = NV

S/C is contained within all sides of this segment. Set visibility parameter to visible and terminate DO loop.

IF
IPL = 0

Set NV to number of sides for this segment.

Determine whether the S/C is contained within the perimeter of any segment of the area target or space volume.

Determine whether S/C is contained within all sides of this segment.

Figure 5-1.- Continued.
Test if S/C was visible previously

IF IVIS > 0

then

Test if S/C was visible previously

IF IAOS > 0

then 30 Increment time

else 31 Generate AOS entry

else

Test if S/C was visible previously

IF IAOS > 0

then 32 Generate LOS entry

else 30 Increment time

Figure 5-1.- Continued.
\[ t = t + \Delta t \]  

No change has occurred. Increment time

31  
\[ TAOS = t \]  
AOS has occurred

Generate AOS entry

\[ IAOS = 1 \]  
\[ t = t + \Delta t \]  
Set AOS parameter to visible and increment time

32  
\[ TLOS = t - \Delta t \]  
LOS has occurred

Generate LOS entry

\[ IAOS = 0 \]  
\[ t = t + \Delta t \]  
Set AOS parameter to not visible and increment time

Figure 5-1.- Concluded.
APPENDIX A

SUBDIVIDING CONCAVE POLYGONS
SUBDIVIDING CONCAVE POLYGONS

This appendix discusses the procedure for subdividing complex concave polygons into two or more simpler convex segments.¹ The purpose of this subdivision process is to permit the equations in section 2 to be used on a segment-by-segment basis to test for containment. The criterion for the S/C (or S/C subsatellite point) to be contained in the concave area target or space volume is that it must be contained in any one of its segments. For convenience, the equations and figures presented in this appendix will use a pentagon as an example.² However, this approach is easily extended to any n-sided polygon.

Figure A-1 presents three examples of concave pentagons. Figure A-1(a) illustrates a pentagon having one concave vertex. Figures A-1(b) and A-1(c) illustrate pentagons having two concave vertices. These pentagons can always be subdivided into triangles by selecting an "appropriate" vertex and connecting nonadjacent vertices (fig. A-2). The maximum number of triangles necessary to completely subdivide any arbitrarily shaped polygon is

\[
N_{\text{seg max}} = n - 2 \quad \text{(A-1)}
\]

where

\[
n = \text{number of sides}
\]

¹A concave polygon is defined to be a polygon that has one or more interior vertex angles exceeding 180 degrees. A convex polygon is defined to be a polygon that has all of its interior angles less than 180 degrees.

²This corresponds to the maximum number of sides specifically addressed in the requirements defined in reference 1.
Figure A-1.- Examples of concave pentagons.

(a) One concave vertex.

(b) Two adjacent concave vertices.

(c) Two nonadjacent concave vertices.
Figure A-2. Examples of subdividing concave pentagons.

(a) One concave vertex.

(b) Two adjacent concave vertices.

(c) Two nonadjacent concave vertices.
Furthermore, the maximum number of interior angles exceeding 180 degrees can also be determined by noting that the sum of the interior angles of the polygon must be equal to the sum of the interior angles of all triangles into which it can be subdivided. Thus, the sum of the interior vertex angles for any arbitrarily shaped polygon is given by

\[
\sum_{i=1}^{n} \gamma_i = (n - 2) \cdot 180 \quad (A-2)
\]

where

\[\gamma_i = \text{interior vertex angles of the polygon}\]

Thus, the maximum number of interior vertex angles exceeding 180 degrees, \(\gamma^*\), is given by

\[
N(\gamma^*)_{\text{max}} = (n - 3) \quad (A-3)
\]

This equation limits the maximum number of concave vertices for a pentagon to two. Figures A-1(b) and A-1(c) illustrate two examples. In figure A-1(b), the two concave vertices are adjacent to each other. In figure A-1(c), the two concave vertices are non-adjacent.

The selection of the "appropriate" vertex to begin the subdivision process is highly dependent upon the shape of the polygon and the number and relationship of the concave vertices. Also, it is not always necessary to subdivide the polygon into triangles. Figure A-3 illustrates another method for subdividing the pentagon of figure A-1(a). In this case, the concave pentagon is subdivided into a four-sided convex polygon and one triangle. Furthermore, figure A-1 by no means exhausts all of the potential pentagon shapes that could be constructed.
Figure A-3.- Alternate subdivision.
Since the shape of the area targets and space volumes will remain static during a mission, it is recommended that the subdivision process be performed manually.\(^1\) There are two distinct advantages to this approach:

a. It eliminates the coding and execution of complex subdivision logic.

b. It can be performed once for each concave area target and space volume and does not have to be repeated each time AOS and LOS times are desired.

The treatment of concave polygons will place additional requirements on the target tables other than those specifically mentioned in reference 1. In addition to the number of sides and coordinates for each vertex, the target tables must also contain the following for each polygon-shaped target

\[ N_{seg} \text{ - number of segments into which the target is subdivided} \]

\[ (3 \geq N_{seg} \geq 1 \text{ for polygons having five or less sides}) \]

\[ V_{Oi} \text{ - integers defining the counterclockwise ordering of the vertices for each segment} \ (i = 1, 2, \ldots, N_{seg}) \]

The use of these additional parameters can best be illustrated by example. For figure A-3, this pentagon is subdivided into two segments. The first segment is a four-sided polygon defined by vertices 1, 2, 3, and 5. The second segment is a triangle defined by vertices 3, 4, and 5. The corresponding parameters for this pentagon would be

\[ N_{seg} = 2 \]

\[ V_{O1} = 1235 \text{ (or 2351 or 3512 or 5123)} \]

\[ V_{O2} = 345 \text{ (or 453 or 534)} \]

\(^1\)This can easily be performed by plotting the vertex points on a Mercator projection.
Similarly, for figures A-2(b) and A-2(c):

a. Figure A-2(b)

\[ N_{\text{seg}} = 3 \]
\[ V_{O_1} = 125 \text{ (or } 251 \text{ or } 512) \]
\[ V_{O_2} = 235 \text{ (or } 352 \text{ or } 523) \]
\[ V_{O_3} = 345 \text{ (or } 453 \text{ or } 534) \]

b. Figure A-2(c)

\[ N_{\text{seg}} = 3 \]
\[ V_{O_1} = 125 \text{ (or } 251 \text{ or } 512) \]
\[ V_{O_2} = 245 \text{ (or } 452 \text{ or } 524) \]
\[ V_{O_3} = 234 \text{ (or } 342 \text{ or } 423) \]

For consistency, this approach can also be used for convex polygons. In this case, \( N_{\text{seg}} \) would be one and \( V_{O_1} \) would be set to the counterclockwise vertex sequence.

For computational purposes, the number of sides for each segment, \( n_i \), can be extracted from the vertex ordering integer, \( V_{O_i} \), as follows

\[ n_i = \text{highest values of } n_i \text{ where } \text{TRUNC} \left( \frac{V_{O_i}}{n_i-1} \right) > 0 \quad (A-4) \]

where

\( \text{TRUNC} \) implies integer truncation.

The vertex numbers, \( V_j \), corresponding to each vertex of the \( i^{th} \) subpolygon can also be extracted from the vertex ordering integer as follows.
\[ V_1 = \text{TRUNC} \left\{ \frac{V_{O_i}}{10^{n_i-1}} \right\} \]  \hspace{1cm} (A-5a)

\[ V_j = \text{TRUNC} \left\{ \frac{V_{O_i} - \sum_{\ell=1}^{j-1} V_{\ell} \cdot 10^{n_i-\ell}}{10^{n_i-j}} \right\} \quad j = 2, 3, \ldots n_i \]  \hspace{1cm} (A-5b)
APPENDIX B

CONIC INTERSECTIONS
CONIC INTERSECTIONS

This appendix presents the equations to compute the intersection points between two right circular cones. The required input parameters are

\[ \hat{V}_1 - \text{unit vector along the axis of the first cone} \]
\[ \gamma_1 - \text{half-cone angle of the first cone} \]
\[ \hat{V}_2 - \text{unit vector along the axis of the second cone} \]
\[ \gamma_2 - \text{half-cone angle of the second cone} \]

The quantities to be computed are

\[ \hat{I}_1 \text{ and } \hat{I}_2 - \text{unit vectors to the intersection points of the two cones} \]

Figure B-1 illustrates the two right circular cones and the intersection points. The coordinate system for the cones is arbitrary and the only restriction is that both \( \hat{V}_1 \) and \( \hat{V}_2 \) must have a common origin and must be expressed in the same system. The first step is to determine whether the relative orientation of the two cones permits any intersections. This can be determined by computing the angle between the axes of the two cones.

\[ \gamma = \cos^{-1} (\hat{V}_1 \cdot \hat{V}_2) \]

These equations can also be used to compute the intersection points between a right circular cone and a plane or the intersection points between two planes.

B-1
Figure B-1.- Intersection of two cones.
The cones will not intersect if

\[ \gamma > \gamma_1 + \gamma_2 \]  \hspace{1cm} \text{[fig. B-2(a)] (B-2)}

or

\[ \gamma + \text{smaller } \{ \gamma_1 : \gamma_2 \} < \text{greater } \{ \gamma_1 : \gamma_2 \} \]  \hspace{1cm} \text{[fig. B-2(b)] (B-3)}

Assuming intersections are possible (i.e., neither equation B-2 nor B-3 is satisfied), the next step is to compute the intersection vectors. Figure B-3 illustrates the geometry. The intersection vectors are symmetrically located on either side of the arc connecting vectors \( \hat{V}_1 \) and \( \hat{V}_2 \). Furthermore, the intersection vectors lie on the perimeter of both cones. Thus, all of the sides of the spherical triangle connecting vectors \( \hat{V}_1 \), \( \hat{V}_2 \), and \( \hat{I}_1 \) are known.

- side a (connecting \( \hat{V}_1 \) and \( \hat{I}_1 \)) = \( \gamma_1 \)
- side b (connecting \( \hat{V}_2 \) and \( \hat{I}_1 \)) = \( \gamma_2 \)
- side c (connecting \( \hat{V}_1 \) and \( \hat{V}_2 \)) = \( \gamma \)

The angle from side 'c' to side 'a' can be determined using the half-angle formula for spherical triangles. The result is

\[ B = 2 \tan^{-1} \left( \frac{K}{\sin(S - \gamma_2)} \right) \]  \hspace{1cm} \text{(B-4)}

where

\[ K = \pm \sqrt{\frac{\sin(S - \gamma_1) \sin(S - \gamma_2) \sin(S - \gamma)}{\sin(S)}} \]  \hspace{1cm} \text{(B-5)}

\[ S = \frac{1}{2} \left( \gamma_1 + \gamma_2 + \gamma \right) \]  \hspace{1cm} \text{(B-6)}
Figure B-2.- Conditions when cones do not intersect.
Figure B-3.- Spherical geometry to compute intersection points.
The dual value of $K$ in equation B-5 produces two values of $B$ in equation B-4. These values of $B$ are equal in magnitude but opposite in sign and correspond to the two intersection points.\footnote{For the special case of $B = 0$, the two intersection vectors are coincident. This physically corresponds to the situation when the two cones are tangent.}

The unit vectors to the two intersection points are given by

\[
\hat{l}_{1,2} = \begin{bmatrix}
\ell_{1x} & \ell_{2x} & V_{lx} \\
\ell_{1y} & \ell_{2y} & V_{ly} \\
\ell_{1z} & \ell_{2z} & V_{lz}
\end{bmatrix}
\begin{bmatrix}
\cos B \sin \gamma_1 \\
+\sin B \sin \gamma_1 \\
\cos \gamma_1
\end{bmatrix}
\]  

(B-7)

where

\[
\hat{V}_1 = (V_{1x}, V_{1y}, V_{1z})
\]

\[
\hat{\ell}_2 = (\ell_{2x}, \ell_{2y}, \ell_{2z})
\]

\[
\hat{\ell}_1 = (\ell_{1x}, \ell_{1y}, \ell_{1z})
\]

(B-8)

\[
\hat{V}_1 \times \hat{V}_2 = \frac{\hat{V}_1 \times \hat{V}_2}{|\hat{V}_1 \times \hat{V}_2|}
\]  

(B-9)
REFERENCE

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