Effects of Fuel-Injector Design on Ultra-Lean Combustion Performance

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Work performed for
U.S. DEPARTMENT OF ENERGY
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ABSTRACT

Emissions data were obtained for six fuel-injector configurations tested with ultra-lean combustion at an inlet-air temperature of 1200 K, a pressure of 250 kPa, and a reference (inlet) velocity of 32 m/s, and a pressure drop of 10% of the total pressure. Fuel injectors included three multiple-source designs and three configurations using a single air-assist injector. Only the multiple-source fuel injectors provided acceptable emissions. Values below the program goals of 16 g CO/kg fuel, 1.9 g HC/kg fuel, and 1.9 g NOx/kg fuel were obtained for the combustion temperature range of 1450 to 1700 K for both a high-blockage (20.2-percent open area) 19-source injector and a low-blockage (41.1-percent open area) 41-source injector. The study showed that high fuel-injector pressure drop may not be required to achieve low-emissions performance at high inlet-air temperatures when the fuel is well-dispersed in the airstream. Further evaluation of both single-source and multiple-source fuel injection systems is required.

INTRODUCTION

This paper presents the results of a study of the effects of fuel-injector design on ultra-lean gas-phase combustion at an inlet-air temperature of 1200 K. The work was performed as supporting research under the DOE Gas Turbine Highway Vehicle Systems Project.

The advanced automotive gas-turbine engine (AGT) currently being developed under DOE contracts will operate with combustor inlet-air temperatures which are significantly higher than those in current gas-turbine engines. Values will typically range from approximately 1100 to 1300 K. Combustor-exit (turbine-inlet) temperatures will be as high as 1650 K. Combustors will be required not only to provide stable combustion but also to operate with low emissions at these conditions.

Low-emissions combustors have traditionally been of the lean premixed type (1-3). Recent combustion tests with inlet-air temperatures of 1100-1250 K (4) concluded, however, that at AGT conditions, premixing will probably be impossible because ignition delay times are too short. That study demonstrated low emissions using a multiple-source fuel injector with a high pressure drop (7 percent at a reference inlet velocity of 32 m/s).

The present study was a preliminary look at whether the multiple-source feature or the high pressure drop is essential to achieving low emissions at high inlet-air temperatures. Emissions of CO, NOx, and UHC were measured for six NASA-designed fuel injector configurations in a 120-mm-diameter flame tube test section. These multiple-source fuel injectors were tested to compare designs with large and small open airflow areas (low and high pressure drops) and different numbers of sources (19 and 41). A single-source air-assist fuel injector was also evaluated with three different duct area restrictions to determine the potential of relatively simple injector configurations. The tests were made with no. 2 diesel fuel. The inlet air was indirectly preheated to 1200 K, the reference velocity was 32 m/s, and the pressure was 250 kPa.

DESCRIPTION OF EXPERIMENT

Test Section

The test section is illustrated in Fig. 1. The test duct was circular in cross section and lined internally with 1.6-mm-thick Fiberfrax sheetmetal liner. Inlet air was indirectly preheated to a temperature of 1200 K. This temperature was monitored with an array of 12 Chromel-Alumel thermocouples located approximately 34 cm upstream of the fuel injector. Inlet temperatures measured near the duct wall were typically less than 40 K below those at the duct centerline.

Fuel Injectors

The fuel-injector concepts tested are shown in Figs. 2(a) to (f). The three multiple-source air-blast fuel injectors were designed by R. Facina at NASA Lewis. They are variations of injectors developed for premixing-prevaporizing-type combustors (5). The single-source injectors were simple air-assist designs constructed at NASA Lewis.

The first multiple-source injector was used in the study of Ref. 4. It had 19 air passages as shown in Fig. 2(a). The objective of this design was to split the airflow into 19 equal flowrates. Equal quantities of fuel entered each passage by flowing through 25-0.5-mm-diameter tubular extensions of the upstream end of each air passage. The air-passages were conical diffusers with a half angle of 7 degrees. The reduced diameter at the plane of fuel injection not only tended to distribute the airflow, but also increased the air velocity to improve the fuel atomization. The fuel drop size was estimated from Lorenzetto and Lefebvre (6) to be about 15 μm for a reference velocity of 32 m/s. The open area seen by the combustion airflow was 20.2 percent of the test-section cross-sectional area (see Table I). At a reference velocity of 32 m/s, the pressure drop of the airflow was 7 percent of the upstream total pressure.

A 19-source fuel injector with 51.7 percent open area was also built to determine if the small open area of the first 19-source injector was necessary to its low-emissions performance. Figure 2(b) describes this fuel injector. The air passages had an entrance diameter of 2.06 mm and a diffuser half angle of 1.4 degrees. The pressure drop was 0.3 percent at 32-m/s reference velocity (Table I). To insure that equal quantities of fuel were discharged into each air passage, 0.5-mm-diameter fuel tubes with a length of 25 cm were again used. Fuel cooling was provided by a flow of shop air in a concentric 0.165-mm-inside diameter tube surrounding each fuel tube. This cooling airflow was about 10 percent of the combustion airflow rate. It discharged into the airstream along with the fuel and would have provided some air assist to help atomize the fuel. The initial fuel drop size was estimated from Ref. 6 to be about 15 μm.

The third injector tested had 41 fuel sources discharging into an equal number of diffusing air passages as shown in Fig. 2(c). Each air-passage entrance was
1.27 cm in diameter with a diffuser half angle of 3.5 degrees. The open airflow area of 41.1 percent resulted in a pressure drop of 0.5 percent at a 32-m/s reference velocity (see Table I). Fuel tubes were 0.069 cm in inside diameter and the concentric air-cooling tubes were 0.155 cm in inside diameter. The correlation of Ref. 6 was again used to predict a duct size of about 15 μm. The cooling air flow was about 15 percent of the combustion airflow rate and provided air assist for fuel atomization.

In addition to tests with these three multiple-source fuel injectors, a single air-assist injector was evaluated. The 19-source fuel injector, a single air-assist injector was evaluated. The first of the three air-assist configurations was mounted in the duct as shown in Fig. 2(d). There was no restriction in the duct passage except for the blockage of the fuel injector tube itself. The open area for the combustion airflow was 96.6 percent and the pressure drop was 0.1 percent of the upstream total pressure at a reference velocity of 32 m/s (Table I).

The remaining two air-assist configurations used a restriction in the duct area so that fuel was injected into a reduced-diameter section. Figure 2(e) shows the 7.5-cm-diameter restriction, and Fig. 2(f) shows the 5-cm-diameter restriction. These restrictions provided flow open areas of 37.8 and 16.3 percent, respectively, and the corresponding pressure drops were 0.7 and 5 percent at a reference velocity of 32 m/s. Three to five cm downstream of the fuel-injection point, the airflow area increased suddenly to the full 12.1-cm diameter of the test duct. This sudden drop of the flow could be expected to assist in flame stabilization.

No igniter was used because combustion reactions occurred spontaneously when fuel was injected into the high-temperature airstream for all fuel injector configurations tested. Combustion products were sampled at a single centerline location approximately 45 cm downstream of the plane of fuel injection. This distance varied for each injector configuration due to slight differences in the hardware used. The sampling probe was water cooled to quench the reactions and freeze the sampled concentrations. A 0.625-cm diameter stainless steel sampling line was electrically heated to maintain a temperature of 400 to 450 K. Through this line, the sampled gases flowed to the emissions analyzers to provide a continuous monitoring of the concentrations of CO, CO₂, unburned hydrocarbons, and nitrogen oxides.

RESULTS AND DISCUSSION

All data reported were obtained with an inlet-air temperature of 1200 K, a pressure of 250 KPa, and a reference (inlet) velocity of 32 m/s.

For this preliminary assessment, emissions were measured only at the duct centerline. This centerline measurement will give an accurate representation of the conditions in the duct only for a homogeneous mixture. An indication of how well the sample represents the conditions in the duct can be obtained by comparing the fuel-air ratio of the sampled gases with the average (metered) fuel-air ratio in the duct. The sample fuel-air ratio is obtained by summing the carbon atoms in the measured CO, CO₂, and UHC emissions.

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lower CO emissions. Subsequent testing resulted in CO emissions very similar to those from the multiple-source injector (see Fig. 3).

The single-air-assist injector with the 7.5-cm-diameter restriction also gave centerline emissions which suddenly changed during the course of testing. CO increased from about 10 g CO/kg fuel at a combustion temperature of 1570 K to 54 g CO/kg fuel at 1615 K (Fig. 3) while NOx decreased from about 37 g NOx/kg fuel to about 23 g NOx/kg fuel (Fig. 4). Emissions then stabilized, defining new curves of emissions vs. temperature. The emissions shifted, opposite to that observed for the single-air-assist injector with no duct restriction, could have been caused by a flow perturbation which moved the start of combustion from a poorly-mixed region to a somewhat more uniform location. For example, combustion may have started initially at the fuel injector but later stabilized at the step just downstream of the fuel injector. The CO for this fuel injector (see Fig. 3) decreased with increasing combustion temperature but was always higher than that observed with the multiple-source injectors for any combustion temperature.

The centerline CO emissions for the air-assist injector with a 5-cm-diameter restriction were about 2 orders of magnitude above the comparable levels for the multiple-source injectors. Only limited testing was performed with this configuration because very high temperature would have been required to obtain much improvement in performance. It is likely that a high-velocity core of reacting flow was created by the duct restrictions used with the air-assist injectors. The residence times in this core would have been significantly less than those of the other multiple-source injectors.Apparently, the fuel-air ratio profile was not sufficiently uniform to vitiate inlet-air temperature of 1200 K, a reference (inlet) velocity of 32 m/s, and a pressure of 250 kPa. Three multiple-source fuel injectors were evaluated to determine the effects of different pressure drops and different numbers of sources (19 and 41). Three single-air-assist injectors were tested as well so that data for fairly simple injectors used the same injector design but different duct-area restrictions to establish if performance was affected by an area reduction at the injection point.

SUMMARY AND CONCLUDING REMARKS

Tests were performed to compare the emissions for six NASA-designed fuel injector configurations at ultra-lean gas-phase combustion conditions in a 12-cm-diameter flame tube (see Fig. 1). The highest velocity core of reacting flow was created by the duct restrictions used with the air-assist injectors. The residence times in this core would have been significantly less than those of the other multiple-source injectors. Apparently, the fuel-air ratio profile was not sufficiently uniform to vitiate inlet-air temperature of 1200 K, a reference (inlet) velocity of 32 m/s, and a pressure of 250 kPa. Three multiple-source fuel injectors were evaluated to determine the effects of different pressure drops and different numbers of sources (19 and 41). Three single-air-assist injectors were tested as well so that data for fairly simple injectors used the same injector design but different duct-area restrictions to establish if performance was affected by an area reduction at the injection point.

Centerline CO and NOx emissions were simultaneously below program goals over a range of combustion temperatures from 1450 to 1700 K for both a 19-source fuel injector with 20-percent open area (high pressure drop) and a 41-source fuel injector with 41-percent open area (low pressure drop). A second 19-source injector with 52 percent open area (low pressure drop) produced significantly higher NOx emissions, apparently because of a less-uniform fuel-air ratio profile.

Three single-source air-assist fuel injector configurations gave generally higher NOx emissions than the multiple-source injectors. Although CO emissions for one single-air-assist injector were acceptable, low CO and NOx could not be achieved simultaneously.

For this study, then, multiple-source fuel injection was required to achieve low emissions ultra-lean combustion while high fuel-injector pressure drop was
not. Single-source injector designs different from those tested in this study need to be evaluated further to determine their potential for low-emissions performance at AGT conditions. Additional tests of multiplesource injectors with a smaller number of sources are also required to determine if simpler designs might provide acceptable performance.

This study demonstrated that the achievement of low emissions with high-inlet-air-temperature combustion is very sensitive to fuel injector design. The development of practical combustors to operate at high inlet-air temperatures will therefore require extensive fuel injector development.

REFERENCES


<table>
<thead>
<tr>
<th>Fuel injector</th>
<th>Open area, percent of duct area</th>
<th>Pressure drop, percent of upstream total pressure, ( V_r, ) m/s</th>
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<tr>
<td>High-pressure-drop 19-source</td>
<td>20.2</td>
<td>7.0 23</td>
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<tr>
<td>Low-pressure-drop 19-source</td>
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<td>0.3 0.9</td>
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<tr>
<td>41-source</td>
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<td>.5 2.2</td>
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<td>Single air-assist with no duct restriction</td>
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<td>.1 .4</td>
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<tr>
<td>Single air-assist with 7.5-cm-diameter restriction</td>
<td>37.8</td>
<td>.7 3</td>
</tr>
<tr>
<td>Single air-assist with 5.5-cm-diameter restriction</td>
<td>16.3</td>
<td>5.0 ---</td>
</tr>
</tbody>
</table>

*\( V_r \) = Reference Velocity, Velocity based on test-section-inlet conditions and 12-cm diameter.
Figure 1. - Test section with high-pressure-drop 19-source fuel injector. (Dimensions are in cm.)
(a) HIGH-PRESSURE-DROP 19-SOURCE FUEL INJECTOR.

Figure 2. Fuel injectors. (Dimensions in cm.)
Figure 2. - Continued.
TYPICAL COOLED FUEL TUBE (ONE OF 41)

FUEL ENTRANCES 4 PLACES

COOLING AIR

COMBUSTION AIR

12 diam

1.27

7.6

44

10 or 11 COOLED FUEL TUBES ENTER FROM EACH OF 4 CIRCUMFERENTIAL LOCATIONS

(c) 41-SOURCE FUEL INJECTOR.

Figure 2 - Continued.
(d) SINGLE-AIR-ASSIST FUEL INJECTOR.

(e) SINGLE-AIR-ASSIST FUEL INJECTOR WITH 7.5-cm-diameter DUCT RESTRICTION.

Figure 2. - Continued.
SHOP AIR

COMBUSTION AIR

FUEL

SHOP AIR

12 diam

5 diam

45°

5

41

NOZZLE TIP DETAILS AS SHOWN IN FIG. 2(d)

(f) SINGLE-AIR-ASSIST FUEL INJECTOR WITH 5-cm-diameter DUCT RESTRICTION.

Figure 2. - Concluded.

Figure 3. - Carbon monoxide emissions at duct centerline. Inlet-air temperature, 1200 K; reference velocity, 32 m/s; pressure, 250 kPa.
Figure 4. - Nitrogen oxides emissions at duct centerline. Inlet-air temperature, 1200 K; reference velocity, 32 m/s; pressure, 250 kPa.

Figure 5. - Trade-off between carbon monoxide and nitrogen oxides emissions at duct centerline. Inlet-air temperature, 1200 K; reference velocity, 32 m/s; pressure, 250 kPa.
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