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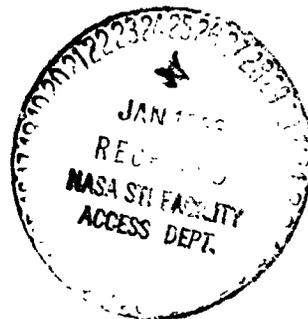
NASA TECHNICAL MEMORANDUM

NASA TM-82504

FY 1982 SCIENTIFIC AND TECHNICAL REPORTS, ARTICLES,
PAPERS, AND PRESENTATIONS

Compiled by Sarah S. Thacker
Management Operations Office

October 1982



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*George C. Marshall Space Flight Center
Marshall Space Flight Center, Alabama*

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FOREWORD

In accordance with the NASA Space Act of 1958 the MSFC has provided for the widest practicable and appropriate dissemination of information concerning its activities and the results thereof.

Since July 1, 1960, when the George C. Marshall Space Flight Center was organized, the reporting of scientific and engineering information has been considered a prime responsibility of the Center. Our credo has been that "research and development work is valuable, but only if its results can be communicated and made understandable to others."

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**GEORGE C. MARSHALL SPACE FLIGHT CENTER
Marshall Space Flight Center, Alabama**

**FY 1982 SCIENTIFIC AND TECHNICAL REPORTS,
ARTICLES, PAPERS, AND PRESENTATIONS**

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TM-82448 November 1981
Spacelab Mission 1 Experiment Descriptions
– Second Edition. Edited by Paul D. Craven.
Space Sciences Laboratory. N82-18234

This document presents brief descriptions of experiments and facilities planned for Spacelab 1. These experiments and facilities were selected from the responses to the Announcement of Opportunity for the first Spacelab mission. The experiments described here have been selected for flight.

This edition supersedes NASA TM-78173, May 1978.

TM-82449 October 1981
Kohoutek Photometric Photography Experiment (S233) Final Report. C. A. Lundquist and P. D. Craven. Space Sciences Laboratory. N82-15007

This report presents the final results of the Skylab 4 experiment S233, Kohoutek Photometric Photography Experiment, which undertook a series of visible light photographs suitable for photometry and for a photographic history of Comet Kohoutek. The report explains the experiment concept, the data reduction method, and the results obtained.

TM-82450 October 1981
Adaptation of the $TH\epsilon\mu$ Formalism for the Analysis of the Equivalence Principle in the Presence of the Weak and Electroweak Interaction. A. J. Fennelly, NAS/NRC Postdoctoral Resident Research Associate. Space Sciences Laboratory. N82-13851

The $TH\epsilon\mu$ formalism, used in analyzing equivalence principle experiments of metric and nonmetric gravity theories, is adapted to the description of the electroweak interaction using the Weinberg-Salam unified $SU(2) \times U(1)$ model. The use of the $TH\epsilon\mu$ formalism is thereby extended to the weak interactions, showing how the gravitational field affects W_{μ}^{\pm} and Z_{μ}^0 boson propagation and the rates of interactions mediated by them. The possibility of a similar extension to the strong interactions via $SU(5)$ grand

unified theories is briefly discussed. Also, using the effects of the potentials on the baryon and lepton wave functions, the effects of gravity on transition rates are determined for β -decay, K-capture, and parity nonconserving transitions mediated in high-A atoms which are electromagnetically forbidden. Three possible experiments to test the equivalence principle in the presence of the weak interactions, which are technologically feasible, are then briefly outlined: (1) K-capture by the ^{55}Fe nucleus (counting the emitted X-rays); (2) forbidden absorption transitions in high-A atoms' vapor; and (3) counting the relative β -decay rates in a suitable α - β decay chain, assuming the strong interactions obey the equivalence principle. The report concludes with an outline of future work concerning (1) Eötvös tests involving the weak-interaction part of nuclear binding energies; (2) (β , γ , β^+) decays of nuclear isotope triplets and (β - γ) angular correlation experiments as weak-interaction equivalence tests; and (3) strong interaction tests.

TM-82451 November 1981
A Quantitative Study of Factors Influencing Lamellar Eutectic Morphology During Solidification. W. F. S. Kaukler, Universities Space Research Association Visiting Scientist. N82-13218

The work was carried out to obtain a quantitative evaluation of the factors that influence the shape of the solid-liquid interface of a lamellar binary eutectic alloy. The experiments were performed using alloys of Carbon Tetrabromide and Hexachloroethane which serve as a transparent analogue of lamellar metallic eutectics. The experimental apparatus that was used was designed to permit direct observation of the solid-liquid interface under very closely controlled conditions. The observed interface shapes were analyzed by computer-aided methods.

The solid-liquid interfacial free energies of each of the individual phases comprising the eutectic system were measured as a function of composition using a "grain boundary groove" technique. The solid-liquid interfacial free energy of the two phases could be evaluated directly

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from the eutectic interface. Various data were measured: the phase diagram for the system, the heat of fusion as a function of composition, and the density as a function of composition.

It was concluded that the shape of the eutectic interface is controlled mainly by the solid-liquid and solid-solid interfacial free energy relationships at the interface and by the temperature gradient present, rather than by interlamellar diffusion in the liquid at the interface, over the range of growth rates studied.

TM-82452 November 1981
An Improved Stress Corrosion Test Medium For Aluminum Alloys. T. S. Humphries and J. E. Coston. Materials and Processes Laboratory. N82-13216

A laboratory test method that is only mildly corrosive to aluminum and discriminating for use in classifying the stress corrosion cracking resistance of aluminum alloys is presented along with the method used in evaluating the media selected for testing. The proposed medium is easier to prepare and less expensive than substitute ocean water.

TM-82453 November 1981
Numerical Stability of an Explicit Finite Difference Scheme for the Solution of Transient Conduction in Composite Media. Warren Campbell. Space Sciences Laboratory. N82-14476

A theoretical evaluation of the stability of an explicit finite difference solution of the transient temperature field in a composite medium is presented. The grid points of the field are assumed uniformly spaced, and media interfaces are either vertical or horizontal and pass through grid points. In addition, perfect contact between different media (infinite interfacial conductance) is assumed. A finite difference form of the conduction equation is not valid at media interfaces; therefore, heat balance forms are derived. These equations were subjected to stability analysis, and a computer graphics code was developed that permitted determination of a maximum time step for a given grid spacing.

TM-82454 December 1981
Environmental Durability of Electroplated Black Chromium. James R. Lowery. Materials and Processes Laboratory. N82-19357

A study was undertaken to determine the durability of nickel-black chromium plated aluminum in an outdoor rural, industrial, and seacoast environment. Test panels were exposed to these environments for 60, 36, and 13 months, respectively. The results of this study showed that no significant optical degradation occurred from exposure to either of these environments, although a considerable amount of corrosion occurred on the panels exposed to the seacoast environment. The rural and industrial atmosphere produced only a slight amount of corrosion on test panels.

TM-82455 January 31, 1982
Longwall Guidance and Control Development: Final Report for the Department of Energy. N82-19637

The Longwall Guidance and Control (G&C) Development Program was begun in 1974 with the aim of determining which systems and subsystems of the longwall system lent themselves to automatic control in the mining of coal. The upper coal/shale interface was identified as the reference for a vertical G&C system, with two sensors (the Natural Background and the sensitized pick) being used to locate and track this boundary. In order to insure a relatively smooth recession surface (roof and floor of the excavated seam), a last and present cut measuring instrument (acoustic sensor) was used. Potentiometers were used to measure elevations of the shearer arms. The integration of these components comprised the Vertical Control System (pitch control). Yaw and roll control were incorporated into a Face Alignment System which was designed to keep the coal face normal to its external boundaries. Numerous tests, in the laboratory and in the field, have confirmed the feasibility of automatic horizon control, as well as determining the face alignment.

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TM-82456 December 1981
Characterization of ASEC BSR 2 Ohm-cm Silicon Solar Cells with Dielectric Wrap-around Contacts as a Function of Temperature and Intensity. A. F. Whitaker and S. A. Little. Materials and Processes Laboratory. N82-19668

Twenty high performance BSE 2 ohm-cm silicon solar cells manufactured by ASEC have been evaluated at 1 AU conditions and at low temperatures and low intensities representative of deep space. These cells showed evidence of series resistance at 1 AU conditions and approximately 50% had reduced power outputs under deep space conditions. Average efficiency of these cells was 12.4% at 1 AU conditions of 1 SC/+25°C.

TM-82457 January 1982
STS-2 Induced Environment Contamination Monitor (IECM) - Quick-Look Report. Edited by E. R. Miller. Space Sciences Laboratory. N82-18870

The STS-2/Induced Environment Contamination Monitor (IECM) mission is described. The IECM system performance is discussed, and IECM mission time events are briefly described. Quick-look analyses are presented for each of the 10 instruments comprising the IECM on the flight of STS-2. Finally, a short summary is presented and plans are discussed for future IECM flights providing more extreme thermal environments, longer on-orbit durations (important for determining offgassing decay rates), and opportunities for direct mapping of Orbiter effluents using the Remote Manipulator System.

TM-82458 December 1981
Propulsion System Ignition Overpressure for the Space Shuttle. R. S. Ryan, J. H. Jones, S. H. Guest, H. G. Struck, M. H. Rheinfurth, and V. S. Verderaime. Systems Dynamics Laboratory. N82-19299

Liquid and solid rocket motor propulsion systems create an overpressure wave during ignition, caused by the accelerating gas particles pushing against or displacing the air contained in the launch pad or launch facility and by the afterburning of the fuel-rich gases. This wave behaves

as a blast or shock wave characterized by a positive triangular-shaped first pulse and a negative half-sine wave second pulse. The pulse travels up the space vehicle and has the potential of either overloading individual elements or exciting overall vehicle dynamics. The latter effect results from the phasing difference of the wave from one side of the vehicle to the other. This overpressure phasing, or ΔP environment, because of its frequency content as well as amplitude, becomes a design driver for certain panels (e.g., thermal shields) and payloads for the Space Shuttle. This paper deals with the history of overpressure effects on the Space Shuttle, the basic overpressure phenomenon, Space Shuttle overpressure environment, scale model overpressure testing, and techniques for suppressing the overpressure environments.

TM-82459 April 1982
Solution Potentials of Several Aluminum Alloys as a Function of Aging Time. Merlin D. Danford. Materials and Processes Laboratory. N82-22345

This report covers the use of solution potentials as a means for determining the tempers of several aluminum alloys. In general, the curves obtained in this study are characterized by a rather sharp drop in solution potential during the first few hours of aging, followed by an inflection or knee, where the shape of the curve changes fairly abruptly and the potentials decrease very slowly with further aging. Times of aging reported in the literature are fairly consistent with the positions of the knees of the curves so obtained, and solution potentials obtained for the various tempers are generally in good agreement with values reported in the literature.

TM-82460 February 1982
Management and Control of Self-Replicating Systems: A Systems Model. Georg von Tiesenhausen. Program Development. N82-25892

In 1980, a conceptual engineering approach to self-replicating systems was achieved by von Tiesenhausen and Freitas. Their designs are based on von Newmann's kinematic version of self-replicating automata. This report expands on the

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systems management and control and the organization of the control elements. After developing the functional requirements of such a system, a hierarchy of three management and control levels is described. These are an autonomous, an external, and an intelligent management and control system. Systems recycling, systems specialization, and information replication are discussed.

TM-82461 January 1982
Space Shuttle STS-1 SRB Damage Investigation Final Report. Clyde D. Nevins. Structures and Propulsion Laboratory.
N82-20235

The physical damage incurred by the Solid Rocket Boosters during reentry on the initial Space Shuttle flight raised the question of whether the hardware, as designed, would yield the low cost per flight desired. An ad hoc committee of technical specialists was chartered to quantify the damage, determine its cause, and recommend specific design changes which would preclude recurrence. Flight data, post-flight analyses, and laboratory hardware examinations were used during the course of the investigation. The resultant findings pointed to two principal causes: (1) failure of the aft skirt thermal curtain at the onset of reentry aerodynamic heating, and (2) overloading of the aft skirt stiffening rings during water impact. Design changes were recommended on both the thermal curtain and the aft skirt structural members to prevent similar damage on future missions.

TM-82462 April 1982
Lagrangian Least-Squares Prediction of Solar Activity. Robert L. Holland, C. A. Rhodes, and Harold C. Fuler, Jr. Space Sciences Laboratory.
N82-27211

This report presents the results of comparison studies on various applications of statistical prediction methods for short-term (months) and long-term (years) forecasting of solar activity. The comparisons indicate that better predictions, in a chi-square sense, are possible by "lining up" the maximum (or minimums, or both) by cycle number. Evidence is also presented to support the existence of an aperiodic variation in the periods as well as the amplitudes.

TM-82463 December 1981
Atmospheric Environment for Space Shuttle (STS-2) Launch. D. L. Johnson and S. C. Brown. Space Sciences Laboratory.
N82-20805

This report presents a summary of selected atmospheric conditions observed near Space Shuttle STS-2 launch time on November 12, 1981, at Kennedy Space Center, Florida. Values of ambient pressure, temperature, moisture, ground winds, visual observations (cloud), and winds aloft are included. The sequence of pre-launch Jimsphere measured vertical wind profiles is given in this report. Also presented are the wind and thermodynamic parameters measured at the surface and aloft in the SRB descent/impact ocean area. Final meteorological tapes, which consist of wind and thermodynamic parameters versus altitude, for STS-2 vehicle ascent and SRB descent have been constructed. The STS-2 ascent meteorological data tape has been constructed by Marshall Space Flight Center in response to Shuttle task agreement No. 989-13-22-368 with Johnson Space Center.

TM-82464 December 1981
Atmospheric Observations for STS-2 Landing. Robert E. Turner, James E. Arnold, and Gregory S. Wilson. Space Sciences Laboratory.
N82-20806

A summary of synoptic weather conditions existing over the western United States is given for the time of Shuttle descent into Edwards Air Force Base, California. The techniques and methods used to furnish synoptic atmospheric data at the surface and aloft for flight verification of the STS-2 Orbiter during its descent into Edwards Air Force Base are specified. Examples of the upper-level data set are given.

TM-82465 January 1982
Statistical Aspects of the 1980 Solar Flares - I. Data Base, Frequency Distributions, and Overview Remarks. Robert M. Wilson. Space Sciences Laboratory.
N82-21135

All 1349 H α flares occurring in 1980 which have known start, maximum brightness, and end times, latitudes, and associated H α importance

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of failed bearing balls and races, and wear track and crack configuration analyses were carried out. In addition, one bearing was tested in laboratory at very high axial loads. The results showed that the cracks were surface initiated and propagated into subsurface locations at relatively small angles. Subsurface cracks were much more extensive than was apparent on the surface. The location of major cracks in the races corresponded to high radial loads rather than high axial loads. There was evidence to suggest that the inner races were heated to elevated temperatures.

A failure scenario was developed based on the above findings. According to this scenario the HPOTP bearings are heated by a combination of high loads and high coefficient of friction (poor lubrication). Very high internal radial loads can be generated by loss of bearing internal clearance resulting from localized heating. These internal radial loads are apparently responsible for the bearing failures. Different methods of extending the HPOTP bearing life are also discussed. These include reduction of axial loads, improvements in bearing design, lubrication and cooling, and use of improved bearing materials.

TM-82471 April 1982
Thermal Monitoring, Measurement, and Control System for a Volatile Condensable Materials (VCM) Test Apparatus. R. E. Ives. Materials and Processes Laboratory.
N82-24472

This report describes a unique thermal monitoring and control concept for a Volatile Condensable Materials (VCM) test apparatus where electric resistance heaters are employed per VCM test specification JSC SP-R-0022A or ASTM standard test method E-595-77. The technique is computer-based, but requires only proportioning ON/OFF relay control signals supplied through a programmable scanner and simple quadrac power controllers.

System uniqueness is derived from automatic temperature measurements and the averaging of these measurements in discrete overlapping temperature zones.

Overall control tolerance proves to be better than $\pm 0.5^{\circ}\text{C}$ from room ambient temperature to 150°C . Using precisely calibrated thermocouples, the method provides excellent temperature control of a small copper VCM heating plate at $125 \pm 0.2^{\circ}\text{C}$ over a 24 hr test period. For purposes of unattended operation, the programmable computer/controller provides a continual data printout of system operation. Real-time operator command is also provided for, as is automatic shutdown of the system and operator alarm in the event of malfunction.

This system has been incorporated into the MSFC Materials and Processes Laboratory VCM Test Facility located in Building 4711.

TM-82472 February 1982
Water Absorption and Desorption in Shuttle Ablator and Insulation Materials. A. F. Whitaker, C. F. Smith, V. A. Wooden, B. E. Cothren, and H. Gregory. Materials and Processes Laboratory.
N82-24299

Shuttle systems ablator and insulation materials have undergone water soak with subsequent water desorption in vacuum. Water accumulation in these materials after a soak for 24 hours ranged from +1.1% for orbiter tile to +161% for SRB MSA-1. After 1 minute in vacuum, water retention ranged from none in the orbiter tile to +70% for SRB cork.

TM-82473 June 1982
Terrestrial Environment (Climatic) Criteria Guidelines for Use in Aerospace Vehicle Development, 1982 Revision. Compiled by Robert E. Turner and C. Kelly Hill.
N82-28317

This document provides guidelines on terrestrial environment data specifically applicable for NASA aerospace vehicles and associated equipment development. The primary geographic areas encompassed are the John F. Kennedy Space Center, Florida; Vandenberg AFB, California; Edwards AFB, California; Michoud Assembly Facility, New Orleans, Louisiana; National Space Technology Laboratory, Bay St. Louis, Mississippi; Lyndon B. Johnson Space Center, Houston, Texas; and the White Sands

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Missile Range, New Mexico. In addition, a section has been included to provide information on the general distribution of natural environmental extremes in the conterminous United States that may be needed to specify design criteria in the transportation of space vehicle subsystems and components. Although not considered as a specific vehicle design criterion, a section on atmospheric attenuation has been added since certain Earth orbital experiment missions are influenced by the Earth's atmosphere. A summary of climatic extremes for worldwide operational needs is also included. This document presents the latest available information on probable climatic extremes and succeeds information presented in TM X-64589, TM X-64757, and TM X-78118. Information is included on atmospheric chemistry, seismic criteria, and on a mathematical model to predict atmospheric dispersion of aerospace engine exhaust cloud rise and growth. There is also a section on atmospheric cloud phenomena. The information in this report is recommended for use in the development of aerospace vehicle and associated equipment design and operational criteria, unless otherwise stated in contract work specifications.

The environmental data in this report are primarily limited to information below 90 km. Environmental criteria for 90 km and above are being documented in a NASA Technical Memorandum entitled "Space and Planetary Environment Criteria Guidelines for Use in Space Vehicle Development (1982 Revision) (report in process).

TM-82474 February 1982
Sunspot Variation and Selected Associated Phenomena: A Look at Solar Cycle 21 and Beyond. Robert M. Wilson. Space Sciences Laboratory. N82-25069

This report gives a brief review of solar sunspot cycles 8 through 21. Mean-time intervals are calculated for maximum-to-maximum, minimum-to-minimum, minimum-to-maximum, and maximum-to-minimum phases for cycles 8 through 20 and 8 through 21. Simple cosine functions with a period of 132 years are compared to, and found to be representative of, the variation of smoothed sunspot numbers at solar maximum (\bar{R}_{MAX}) and minimum (\bar{R}_{MIN}). A comparison

of cycles 20 and 21 is given, leading to a projection for activity levels during the Spacelab 2 era (tentatively, November 1984). A prediction is made for cycle 22 (i.e., occurrence dates of \bar{R}_{MIN} and \bar{R}_{MAX} and their corresponding values). Major flares (importance class ≥ 1) are observed to peak several months subsequent to \bar{R}_{MAX} during cycle 21 and to be at minimum level several months after \bar{R}_{MIN} . Additional remarks are given for flares, gradual-rise-and-fall (GRF) radio events and 2800-MHz radio emission. The major thrust of this report is to estimate certain solar activity parameters, especially as they relate to the near-term Spacelab 2 time frame. This report should not be construed to represent a detailed, highly accurate, predictive scheme.

TM-82475 February 1982
Statistical Aspects of the 1980 Solar Flares: II. Solar Cycle Activity Relationships and Additional Remarks. Robert M. Wilson. Space Sciences Laboratory. N82-25070

Based on 1349 H α flares with X-ray counterparts, an investigation into the relationship between rise time, decay time, duration, latitude, H α importance, and X-ray class with 2800-MHz radio emission (F_{2800}) has been accomplished. An important finding is that during 1980 both the number of H α importance class 1 and number of X-ray class M (and M+X) flares appeared to be rather strongly related to F_{2800} , in a positive sense; i.e., number of class 1 and class M events increased as F_{2800} increased. This is the second part of a three-part study of the 1980 solar flares. This series gives associational aspects as related to flares occurring in 1980. No effort has been made to model flare frequency correlation and distribution based on more advanced statistical techniques.

TM-82476 March 1982
The Marshall Space Flight Center KC-135 Zero Gravity Test Program for FY1981. Edited by R. E. Shurney. Systems Analysis and Integration Laboratory. N82-26350

During FY81, researchers from Marshall Space Flight Center (MSFC) conducted seven

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separate investigations during 23.5 hours of testing aboard the KC-135 zero-gravity aircraft, based at Ellington Air Force Base, Texas. Although this represented fewer hours than initially projected, all experiment objectives were met or exceeded. This Technical Memorandum compiles all results achieved by MSFC users during FY81, a year universally considered to be highly productive.

We thank the Aircraft Operations people at Johnson Space Center for their enthusiastic support, this year and in years past.

TM-82477 January 1982
Spacelab Mission 2 Experiment Descriptions - Second Edition. Edited by K. Stuart Clifton. Space Sciences Laboratory.

A brief description is presented of the Spacelab 2 Mission and the 12 multidisciplinary experiments selected to fly on board. These experiments include the following: Vitamin D Metabolites and Bone Demineralization, Interaction of Oxygen and Gravity Influenced Lignification, Eye Plasma Diagnostics Package, Plasma Diagnostic Experiments for Ionospheric and Radio Cosmical Studies, Small Helium-Cooled IR Telescope, Elemental Composition and Energy Spectra of Cosmic Ray Nuclei, Hard X-Ray Imaging of Clusters of Galaxies and Other Extended X-Ray Sources, Solar Magnetic and Velocity Field Measurement System, Solar Coronal Helium Abundance Spacelab Experiment, Solar UV High Resolution Telescope and Spectrograph, Solar UV Spectral Irradiance Monitor, and Properties of Superfluid Helium in Zero-G. This report supersedes NASA TM-78198.

TM-82479 May 1982
Computerized Data Collection and Reduction From an X-Ray Diffractometer. John C. McClure. Materials and Processes Laboratory. N82-25809

A series of computer programs has been written for use with a Philips X-ray Diffractometer and a Hewlett Packard 9825A Desk Top Computer. These programs permit the collection and storage on disk of the number of X-ray

counts and the associated 2-theta angles across line profiles. Automatic background subtraction, integrated intensity, correction for the angular dependence of the Lorentz, polarization, and atomic scattering factors, peak location, K-alpha 2 removal, and calculation of Fourier coefficients are performed. This technical note is documentation for these programs and should provide a guide to their use. The programs are written in HPL which is a Hewlett Packard variation of BASIC. The programs are written for a computer configured with a disk drive, but they can be easily modified to run from the cassette tape drive that is integral to the 9825A Computer.

Copies of these programs are available on cassette from the author.

TM-82480 April 1982
Atmospheric Environment for Space Shuttle (STS-3) Launch. D. L. Johnson, S. C. Brown, and G. W. Batts. Space Sciences Laboratory. N82-27913

This report presents a summary of selected atmospheric conditions observed near Space Shuttle STS-3 launch time on March 22, 1982, at Kennedy Space Center, Florida. Values of ambient pressure, temperature, moisture, ground winds, visual observations (cloud), and winds aloft are included. The sequence of pre-launch Jimsphere measured vertical wind profiles is given in this report. Also presented are the wind and thermodynamic parameters measured at the surface and aloft in the SRB descent/impact ocean area. Final meteorological tapes, which consist of wind and thermodynamic parameters versus altitude, for STS-3 vehicle ascent and SRB descent have been constructed. The STS-3 ascent meteorological data tape has been constructed by Marshall Space Flight Center in response to Shuttle task agreement No. 989-13-22-368 with Johnson Space Center.

TM-82481 June 1982
Atmospheric Observations for STS-3 Landing. Robert E. Turner, James E. Arnold, Gregory S. Wilson, and Wade Batts. Space Sciences Laboratory. N82-29827

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A summary of synoptic weather conditions existing over the western United States is given for the time of Shuttle descent into White Sands Missile Range, New Mexico. The techniques and methods used to furnish synoptic atmospheric data at the surface and aloft for flight verification of the STS-3 Orbiter during its descent into White Sands Missile Range are specified. Examples of the upper-level data set are given.

TM-82482 April 1982
The Human Role In Space. Stephen B. Hall, George von Tiesenhausen, and Gary W. Johnson. Program Development.
N82-27987

This report describes a limited Marshall Space Flight Center in-house study on the human role in space. This study was performed during 1980 and its procedures and results were only available in chart form. Since the methodology and findings could be of interest to a larger circle of people the report form was chosen as an efficient way to disseminate the study results for future reference. It should be noted that the mission model used in this study has changed; however, the approach taken and the general conclusions have remained valid.

TM-82483 July 1982
A Bivariate Gamma Probability Distribution with Application to Gust Modeling. O. E. Smith, S. I. Adelfang, and J. D. Tubbs. Space Sciences Laboratory. N82-29094

A five-parameter gamma distribution (BGD) having two shape parameters, two location parameters, and a correlation parameter is investigated. This general BGD is expressed as a double series and as a single series of the modified Bessel function. This general BGD reduces to the known special case for equal shape parameters. Practical functions for computer evaluations for the general BGD and for special cases are presented. Applications of the general BGD are to be found in reliability theory, signal noise, and meteorology. In this paper, applications to wind gust modeling for the ascent flight of the Space Shuttle are illustrated.

TM-82484 May 1982
Instrument Manual for the Retarding Ion Mass Spectrometer on Dynamics Explorer-1. S. A. Fields, C. R. Baugher, C. R. Chappell, D. L. Reasoner, H. D. Hammack, W. W. Wright, and J. H. Hoffman. Space Sciences Laboratory. N82-30527

The Retarding Ion Mass Spectrometer (RIMS) for Dynamics Explorer-1 is an instrument designed to measure the details of the thermal plasma distribution. It combines the ion temperature determining capability of the retarding potential analyzer with the compositional capabilities of the mass spectrometer and adds multiple sensor heads to sample all directions relative to the spacecraft ram direction. This manual provides a functional description of the RIMS, the instrument calibration, and a description of the commands which can be stored in the instrument logic to control its operation.

TM-82485 June 1982
Space Telescope Neutral Buoyancy Simulations - The First Two Years. Fred G. Sanders. Systems Analysis and Integration Laboratory. N82-30942

This brief illustrated report of neutral buoyancy simulations conducted to validate the crew systems interface as it relates to space telescope on-orbit maintenance and contingency operations begins with the initial concept validation tests using low-fidelity mockups in August 1979, and progresses through the entire spectrum of proposed space telescope refurbishment and selected contingencies using upgraded mockups which reflect flight hardware as of August 1981. It contains findings which may be applicable to future efforts of a similar nature.

TM-82486 June 1982
Phase Linear Interferometer Experiment Maintenance and Calibration Manual. Steven J. Goodman. Space Sciences Laboratory. N82-29581

The Phase Linear-Interferometer Experiment (PLIE) Maintenance and Calibration Manual describes the necessary procedures for assuring continuous lightning sferics data collection at

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Marshall Space Flight Center. A sister station is operating continuously at Southwest Research Institute. The PLIE is being evaluated as a candidate RF sensor to support the space-based optical lightning mapper system.

TM-82487 May 1982
An Analytical Approach to Thermal Modeling of Bridgman-Type Crystal Growth: One-Dimensional Analysis, Computer Program Users Manual. Ernestine Cothran. Space Sciences Laboratory. N82-30106

This work documents the computer program written in support of R. J. Naumann's "An Analytical Approach to Thermal Modeling of Bridgman-Type Crystal Growth: One-Dimensional Analysis" (accepted for publication in The Journal of Crystal Growth 1982).

The program listing and flow charts are included, along with the complete thermal model. Sample problems include detailed comments on input and output to aid the first-time user.

This report will be of particular value to the scientific community desiring a one-dimensional analysis of crystal growth to guide more complicated numerical analysis.

TM-82488 May 1982
Torques on the Gyro in the Gyro Relativity Experiment. Peter Eby. Space Sciences Laboratory. N82-30526

A discussion and calculation of all the important torques on the Gyro in the Gyro Relativity Experiment is given. This is a similar but independent analysis from that of Everitt. It reaches the same conclusions but includes many of the details left out in previous documentation. It also extends previous analysis of electrical torques. The report gives the conditions under which gyro drifts can be kept well below the relativistic effects predicted by General Relativity.

TM-82489 June 1982
STS-3 Induced Environment Contamination Monitor (IECM) - Quick-Look Report.

Edited by E. R. Miller and J. A. Fountain.
Space Sciences Laboratory. N82-31825

The STS-3/Induced Environment Contamination Monitor (IECM) mission is described. The IECM system performance is discussed, and IECM mission time events are briefly described. Quick-look analyses are presented for each of the 10 instruments comprising the IECM on the flight of STS-3. Finally, a short summary is presented and plans are discussed for future IECM flights, and opportunities for direct mapping of Orbiter effluents using the Remote Manipulator System.

TM-82490 June 1982
STS-3 Main Parachute Failure. Roy Runkle and Keith Henson. Structures and Propulsion Laboratory. N82-29349

March 22, 1982, at 11:00 a.m. Eastern Standard Time (EST), the third launch of the United States Space Shuttle (STS-3) took place. During the reentry phase of the two Solid Rocket Boosters (SRBs), one 115-ft diameter main parachute failed on the right-hand SRB (A12). This parachute caused the SRB to impact the ocean at 110 ft/sec in lieu of the expected "3 parachute" impact velocity of 88 ft/sec. This higher impact velocity relates directly to more SRB aft skirt and more motor case damage. A parachute failure team was formed to assess the cause of the parachute failure, the potential risks of losing an SRB as a result of this failure, and to recommend fixes to ensure that the probability of chute failures of this type in the future will be low. The team's members were from Marshall Space Flight Center, the parachute subsystem contractor, and industry-recognized parachute experts from Sandia Laboratories.

TM-82491 July 1982
Shuttle VLBI Experiment: Technical Working Group Summary Report. Samuel H. Morgan and David H. Roberts (editors). Advanced Systems Office.

This report provides a quantitative description of the gain in interferometric resolution of extragalactic sources at radio frequencies which can be achieved by placing a Very Long Baseline Interferometry (VLBI) antenna in space. The

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report describes in some detail a VLBI demonstration experiment using a large deployable antenna, which will, if realized, be a very acceptable first venture for VLBI in space. The material presented in this report was compiled by a Shuttle VLBI Experiment Working Group which was chartered in 1981 by the Office of Space Science and Applications to develop the rationale and a technical plan for the experiment. The report also includes a tutorial on VLBI, a summary of the technology available for the experiment and a preliminary mission scenario.

TM-82493 July 1982
Variable Reluctance Proximity Sensors for Cryogenic Valve Position Indication. R. A. Cloyd. Structures and Propulsion Laboratory.

This test was conducted to determine the performance of a variable reluctance proximity sensor system when installed in an External Tank vent/relief valve. The sensors were used as position indicators. The valve and sensors were cycled through a series of thermal transients; while the valve was being opened and closed pneumatically, the sensor's performance was being monitored. During these thermal transients, the vent valve was cooled 10 times by liquid nitrogen and 2 times by liquid hydrogen. It was concluded that the sensors were acceptable replacements for the existing mechanical switches. However, the sensors need a mechanical override for the target similar to what is presently used with the mechanical switches. This override could insure contact between sensor and target and eliminate any problems of actuation gap growth caused by thermal gradients.

TM-82494 June 1982
Optical Studies of a Model Binary Miscibility Gap System. L. L. Lacy, W. K. Witherow, B. R. Facemire, and G. M. Nishioka. Space Science Laboratory.

In order to develop a better understanding of separation processes in binary miscibility gap metal alloys, model transparent fluid systems are being studied. The system selected was diethylene glycol-ethyl salicylate (DEG/ES) which has convenient working temperatures (288-350K), low

toxicity, and is relatively easy to purify. The system is well characterized with respect to its phase diagram, density, surface and interfacial tensions, viscosity and other pertinent physical properties. Studies of migration of the dispersed phase in a thermal gradient were performed using conventional photo microscopy. Velocities of the droplets of the dispersed phase were measured and compared to calculated rates which included both Stokes and thermal components. A holographic microscopy system was used to study growth, coalescence, and particle motions. Sequential holograms allowed determination of particle size distribution changes with respect to time and temperature. Holographic microscopy is capable of recording particle densities up to 10^7 particles/cm³ and is able to resolve particles of the order of 2 to 3 μ m in diameter throughout the entire volume of the test cell. Holography offers advantages over other optical techniques. The reconstructed hologram produces a wavefront that is identical to the original wavefront as it existed when the hologram was made. The reconstructed wavefront is analyzed using a variety of conventional optical methods.

TM-82495 July 1982
Atmospheric Observations for STS-4 Landing. Robert E. Turner, James E. Arnold, and Wade Batts. Space Science Laboratory.

A summary of synoptic weather conditions existing over the western United States is given for the time of Shuttle descent into Edwards Air Force Base, California. The techniques and methods used to furnish synoptic atmospheric data at the surface and aloft for flight verification of the STS-4 Orbiter during its descent into Edwards Air Force Base are specified. Examples of the upper-level data set are given.

TM-82496 September 1982
Materials Processing in Space Program Tasks. Compiled by Elizabeth Pentecost. Space Science Laboratory.

This report is a compilation of the active research tasks as of the end of the fiscal year 1982 of the Materials Processing in Space Program, NASA-Office of Space and Terrestrial Applications, involving several NASA centers and

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other organizations. The purpose of this document is to provide an overview of the program scope for managers and scientists in industry, university, and government communities. The report is structured to include an introductory description of the program, its history, strategy and overall goal; identification of the organizational structures and people involved; and a description of each research task, together with a list of recent publications.

The tasks are grouped into four categories: Crystal Growth; Solidification of Metals, Alloys, and Composites; Fluids, Transports, and Chemical Processes, and Ultrahigh Vacuum and Containerless Processing Technologies.

TM-82497 August 1982
Tolerance Requirements to Prevent Fluid Leakage in the Crucible/Plunger MEA Experiment MPS 770030. Thomas J. Rathz. Space Science Laboratory.

The work described was motivated by the unexpected leakage of molten Al-In out of the crucible of a proposed MEA materials processing in space experiment. The molten metals use a spring-loaded plunger to eliminate most free surfaces. The critical criteria necessary to initiate flow and the rate of fluid flow into the crucible/plunger annulus is calculated. Experimental in situ X-radiographs are interpreted according to the calculations. Also presented is a short note on possible effects of capillary flow if wetting occurs between crucible/plunger and liquids.

TM-82498 July 1982
Atmospheric Environment for Space Shuttle (STS-4) Launch. D. L. Johnson, C. K. Hill, and G. W. Batts. Space Science Laboratory.

This report presents a summary of selected atmospheric conditions observed near Space Shuttle STS-4 launch time on June 27, 1982, at Kennedy Space Center, Florida. Values of ambient pressure, temperature, moisture, ground winds, visual observations (cloud), and winds aloft are included. The sequence of prelaunch Jimsphere measured vertical wind profiles is given in this report. Also presented are the wind and thermodynamic parameters measured at the surface and aloft in the SRB descent/impact

ocean area. Final meteorological tapes, which consist of wind and thermodynamic parameters versus altitude, for STS-4 vehicle ascent and SRB descent have been constructed. The STS-4 ascent meteorological data tape has been constructed by Marshall Space Flight Center in response to Shuttle task agreement No. 989-13-22-368 with Johnson Space Center.

TM-82499 July 1982
Refurbishment of SRB Aluminum Components by Walnut Hull Blast Removal of Protective Coatings. Wendell R. Colberg, Gail H. Gordon, and Charles H. Jackson. Materials and Processes Laboratory.

A test program was conducted to develop, optimize, and scale-up an abrasive blasting procedure for refurbishment of specific SRB components: Aft Skirt, Forward Skirt, Frustrum, and painted piece parts.

Test specimens utilizing 2219 T87 aluminum substrate of varying thicknesses were prepared and blasted at progressively increasing pressures ($2.76 \times 10^5 - 5.52 \times 10^5 \text{ N/m}^2$) with selected abrasives. Specimens were then analyzed for material response. The optimum blasting parameters were determined on panel specimens and verified on a large cylindrical Integrated Test Bed (ITB). This report presents findings and conclusions of that study.

TM-82500 September 1982
Exothermic Furnace Module Development. Roy R. Darnell and Richard M. Poorman. Materials Processing In Space Projects Office.

An Exothermic Furnace Module (EFM) has been developed to rapidly heat and cool a 0.820-in. (2.1 cm) diameter by 2.75-in. (7.0 cm) long TZM Molybdenum alloy crucible. The crucible contains copper, oxygen, and carbon for processing in a low-g environment. Peak temperatures of 1270°C were obtainable 3.5 min after start of ignition, with cooling below 950°C some 4.5 min later. These time-temperature relationships were conditioned for a Foam-Copper Experiment, Space Processing Applications Rocket (SPAR) Experiment 77-9, in a sounding rocket having a low-g period of 5 min.

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TP-1932 November 1981
Space Shuttle Main Engine Controller.
Russell M. Mattox and J. B. White. Data
Systems Laboratory. N82-11784

A technical description of the Space Shuttle Main Engine Controller, which provides engine checkout prior to launch, engine control and monitoring during launch, and engine safing and monitoring in orbit, is presented. Each of the major controller subassemblies, the central processing unit, the computer interface electronics, the input electronics, the output electronics, and the power supplies are described and discussed in detail along with engine and orbiter interfaces and operational requirements.

The controller represents a unique application of digital concepts, techniques, and technology in monitoring, managing, and controlling a high performance rocket engine propulsion system. The operational requirements placed on the controller, the extremely harsh operating environment to which it is exposed, and the reliability demanded, result in the most complex and ruggedized digital system ever designed, fabricated, and flown.

TP-1933 November 1981
The Aerodynamics of Bodies in a Rarefied Ionized Gas With Applications to Spacecraft Environmental Dynamics. Nobie H. Stone. Space Sciences Laboratory.
N82-15116

This study consists of two parts: an experimental parametric investigation and an in-depth critical review of knowledge in the field derived from previous experimental investigations, theoretical treatments, and ionospheric satellite data. The objectives are to provide a parametric description of the electrostatic interaction of a mesosonic, collisionless plasma with conducting bodies on the order of 1 to 10 Debye lengths in size, and to extend this description to the satellite-ionospheric interaction, where possible.

New experimental findings include: (1) converging ion streams in the near wake whose inclination to the wake axis and crossing point loca-

tion depend on Φ_b and $(SR_d^{0.24}/|\Phi_b|^{1/2})$, respectively, where Φ_b is the normalized body potential, S , the ion acoustic Mach number, and R_d , the Debye ratio; (2) that two mechanisms with different Φ_b dependences create the mid-wake axial ion peak whose maximum amplitude and width depend on $[S/|\Phi_b|^{1/2}]$ and $|\Phi_b|^{-1/2}$, respectively; (3) the morphology and amplitude of the axial ion peak depend on the geometry of the plasma sheath, which varies with thickness (and therefore R_d and Φ_b) for bodies with square cross sections, but is independent of thickness for spherical and long cylindrical bodies; (4) the wake of the geometrically complex body appears to be a linear superposition of the wakes of its simple geometric components; (5) previously observed electron heating may be explained by a wave-particle interaction resulting from a two-stream instability produced by fast, plasma stream ions passing through slow, charge exchange ions; and (6) vector ion flux measurements show converging ion streams at the wake axis and direct evidence of ion streams deflected from the wake axis by the positive space charge potential associated with the axial ion peak.

The extension to the satellite-ionospheric interaction utilizes qualitative scaling and indicates that similar, but smaller amplitude, wake structures may be expected for small or highly charged bodies. However, for large bodies at small potentials, the structure may be diffused by the thermal ion motion and the dispersion resulting for space charge potentials.

TP-1935 November 1981
Concept for a Power System Controller for Large Space Electrical Power Systems. Louis F. Lollar, John R. Lanier, Jr., and James R. Graves. Electronics and Control Laboratory.
N82-11109

A need for autonomous control of large electrical power systems has emerged. A Marshall Space Flight Center Director's Discretionary Fund task is undertaken to develop technology for a fail-operational Power System Controller (PSC) utilizing microprocessor technology for managing the

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distribution and power processor subsystems of a large multi-kW space Electrical Power System. The task involved determining the specific functions which must be performed by the PSC, determining the best microprocessor available to do the job, and determining the feasibility, cost savings, and applications of a PSC. A limited function breadboard version of a PSC was developed to demonstrate the concept and potential cost savings.

TP-1949 December 1981
Systems Analysis Approach to Deriving Design Criteria (Loads) for Space Shuttle and Its Payloads, Volume I – General Statement of Approach. Robert S. Ryan, Tulon Bullock, Wayne B. Holland, Dennis A. Kross, and Larry A. Kiefling. Systems Dynamics Laboratory. N82-14203

Derivation of a set of design loads criteria for a space system that provides a specified launch or operational probability, adequate lifetime, and safety factors and, at the same time, meet low-cost, high-performance (low weight in general) requirements is the major problem facing engineering and program personnel. Stated another way, how do you achieve an optimized design from the system standpoint under the low-cost, high risk constraints of the present day environment? The answer to this question is compounded by the complex mission models and structural configurations which have strong interaction or coupling between structures, control, propulsion, thermal, aeroelastic, and performance. Basic to this question is how to treat vehicle system parameters and environment uncertainties. Space Shuttle, the most complex transportation system designed to date, illustrates the requirement for an analysis approach that considers all major disciplines simultaneously. Its unique cross coupling and high sensitivity to aerodynamic uncertainties and high performance requirements dictated a less conservative approach than those taken in prior programs. Analyses performed for the Space Shuttle and certain payloads, Space Telescope and Spacelab, are used as examples in Volume 2. These illustrate the requirements for system analysis approaches and criteria, including dynamic modeling requirements, test require-

requirements, control requirements, and the resulting design verification approaches. A survey of the problem, potential approaches available as solutions, implications for future systems, and projected technology development areas are addressed in this report. This report is divided into two independent volumes. Volume 1 deals with the philosophy and general loads analysis approaches. Volume 2 gives the Shuttle examples. Readers can read both or choose either, since they are written to be independent.

TP-1950 December 1981
System Analysis Approach to Deriving Design Criteria (Loads) for Space Shuttle and Its Payloads, Volume II – Typical Examples. Robert S. Ryan, Tulon Bullock, Wayne B. Holland, Dennis A. Kross, and Larry A. Kiefling. Systems Dynamics Laboratory. N82-15106

Derivation of a set of design loads criteria for a space system that provides a specified launch or operational probability, adequate lifetime, and safety factors and, at the same time, meet low-cost, high-performance (low weight in general) requirements is the major problem facing engineering and program personnel. Stated another way, how do you achieve an optimized design from the system standpoint under the low-cost, high risk constraints of the present day environment? The answer to this question is compounded by the complex mission models and structural configurations which have strong interaction or coupling between structures, control, propulsion, thermal, aeroelastic, and performance. Basic to this question is how to treat vehicle system parameters and environment uncertainties. Space Shuttle, the most complex transportation system designed to date, illustrates the requirement for an analysis approach that considers all major disciplines simultaneously. Its unique cross coupling and high sensitivity to aerodynamic uncertainties and high performance requirements dictated a less conservative approach than those taken in prior programs. Analyses performed for the Space Shuttle and certain payloads, Space Telescope and Spacelab, are used as examples in Volume 2. These illustrate the requirements for system analysis approaches and criteria, including

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dynamic modeling requirements, test requirements, control requirements, and the resulting design verification approaches. A survey of the problem, potential approaches available as solutions, implications for future systems, and projected technology development areas are addressed in this report. This report is divided into two independent volumes. Volume 1 deals with the philosophy and general loads analysis approaches. Volume 2 gives the Shuttle examples. Readers can read both or choose either, since they are written to be independent.

TP-1987 March 1982
Damping Seals for Turbomachinery. George L. von Pragenau. Systems Dynamics Laboratory. N82-20183

A rotor seal is proposed that restricts leakage like a labyrinth seal, but extends the stabilizing speed range beyond twice the first critical speed. The dynamic parameters are derived from bulk flow equations without requiring a dominant axial flow. The flow is considered incompressible and turbulent. Damping seals are shown to be feasible for extending the speed range of high performance turbomachinery beyond the limit imposed by conventional seals.

TP-1988 March 1982
Sensitivity Analysis of the Space Shuttle to Ascent Wind Profiles. Orvel E. Smith and Lambert D. Austin, Jr. Space Sciences Laboratory. N82-20236

This report presents a parametric sensitivity analysis of the Space Shuttle ascent flight to the wind profile. Engineering systems parameters are obtained by flight simulations using wind profile models and samples of detailed (Jimsphere) wind profile measurements. The wind models used are the synthetic vector wind model, with and without the design gust, and a model of the vector wind change with respect to time. From these comparison analyses an insight is gained on the contribution of winds to ascent subsystems flight parameters.

TP-1998 December 1981
Application of a Computerized Vibroacoustic Data Bank for Random Vibration Criteria

Development. Robin C. Ferebee. Systems Dynamics Laboratory. N82-20238

A computerized data bank system has been developed for utilization of large amounts of vibration and acoustic data to formulate component random vibration design and test criteria. This system consists of a computer, graphics tablets, and a dry-silver hard copier which are all desk-top type hardware and occupy minimal space. Currently, the data bank contains data from the Saturn V and Titan III flight and static test programs. The vibration and acoustic data are stored in the form of power spectral density and one-third octave band plots over the frequency range from 20 to 2000 Hz. The data were stored by digitizing each spectral plot by tracing with the graphics tablet. The digitized data were statistically analyzed, and the resulting 97.5 percent confidence levels were stored on tape along with the appropriate structural parameters. Standard extrapolation procedures were programmed for prediction of component random vibration test criteria for new launch vehicle and payload configurations.

A user's manual is included to guide potential users through the programs.

TP-2045 April 1982
A Stability Analysis of AVE-IV Severe Weather Soundings. Dale L. Johnson. Space Sciences Laboratory.

An investigation was made to determine whether the stability and vertical structure of an average severe storm sounding, consisting of both thermodynamic and wind vertical profiles, could be distinguished from an average lag sounding taken 3 to 6 hours prior to severe weather occurrence. The term "average" is defined here to indicate the arithmetic mean of a parameter, as a function of altitude, determined from a large number of available observations taken either close to severe weather occurrence, or else more than 3 hours before it occurs. The investigative computations were also done to help determine if a severe storm forecast scheme or index could possibly be used or developed.

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The study presents these mean vertical profiles of thermodynamic and wind parameters as a function of severity of the weather, determined from manually digitized radar (MDR) categories observed during the National Aeronautics and Space Administration (NASA) Atmospheric Variability Experiment IV (AVE-IV) which took place on April 24-25, 1975. Profile differences and stability index differences are presented along with the development of the Johnson Lag Index (JLI) which is determined entirely upon environmental vertical parameter differences between conditions 3 hours prior to severe weather, and severe weather itself.

All of the stability indices tested were then used on a separate and independent data sample (AVE-SESAME-I) consisting of individual soundings taken during April 10-11, 1979. The AVE-SESAME-I data profiles are presented along with stability index computations for each. All of the stability indices tested appeared to do a reasonable job in indicating both the severe weather as well as the nonsevere weather environment. As a pre-severe weather lag (3 to 6 hours) index, only the JLI appears to show promise as a potential forecast index. More testing of this index, however, is needed.

TP-2069 August 1982
Atmospheric Constraint Statistics for the Space Shuttle Mission Planning. O. E. Smith, G. W. Batts, and J. A. Willett. Space Sciences Laboratory.

This report presents the procedures used to establish statistics of atmospheric constraints of interest to the Space Shuttle mission planning. The statistics considered are for the frequency of occurrence, runs, and time conditional probabilities of several atmospheric constraints for each of the Space Shuttle mission phases. The mission phases considered are (1) prelaunch, (2) launch, (3) return to launch site, (4) abort once around landing, and (5) end of mission landing.

TP-2086 June 1982
Nonlinear Optimization with Linear Constraints Using a Projection Method. Thomas Fox. Systems Dynamics Laboratory.

This report examines and discussed nonlinear optimization problems that are encountered in science and industry. A new method of projecting the gradient vector onto a set of linear constraints is developed, and a program that uses this method is presented. The algorithm that generates this new projection matrix is based on the Gram-Schmidt method and overcomes some of the objections to the Rosen projection method. This should make the projection method of optimization with linear constraints more attractive to users.

TP-2089 July 1982
The Pinhole/Occluder Facility - Executive Summary. J. R. Dabbs, E. A. Tandberg-Hanssen, and H. S. Hudson.

The outer solar atmosphere exhibits a great variety of dynamic and energetic plasma phenomena, from the catastrophic energy release of solar flares to the steady acceleration of the solar wind. Observations from space in the past two maxima of the solar activity cycle have more than whetted the appetite for understanding the physics of the solar corona. The Pinhole/Occluder Facility contains the instruments necessary for achieving fuller knowledge: broad-band X-ray imaging, combined with simultaneous ultraviolet and white-light spectroscopy and imaging.

X-ray astronomy has progressed, through the surveys by small satellites and the "deep" observations of soft X-rays by the Einstein Observatory, to a level at which it has become a major component of astronomical investigation. The Pinhole/Occluder represents the first serious effort to broaden the spectral band available to X-ray astronomers at high angular resolution (below one arc second), and it is thus an effective complement to AXAF and other future soft X-ray facilities.

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The information in this report has been reviewed for technical content. Review of any information concerning Department of Defense or Atomic Energy Commission programs has been made by the MSFC Security Classification Officer. This report, in its entirety, has been determined to be unclassified.



J. N. FOSTER
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and Program Support

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