Research on Aviation Fuel Instability

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Current aircraft turbine fuels do not present a significant problem with fuel thermal stability. However, turbine fuels with broadened properties or nonpetroleum-derived fuels may have reduced thermal stability because of their higher content of olefins, heteroatoms, and trace metals. Moreover, advanced turbine engines will increase the thermal stress on fuels because of their higher pressure ratios and combustion temperature.

In recognition of the importance of this problem, NASA Lewis is currently engaged in a broadly-based research effort to better understand the underlying causes of fuel thermal degradation. The progress and status of our various activities in this area will be discussed in this paper. Topics to be covered include: nature of fuel instability and its temperature dependence, methods of measuring the instability, chemical mechanisms involved in deposit formation, and instrumental methods for characterizing fuel deposits. Finally, some preliminary thoughts on design approaches for minimizing the effects of lowered thermal stability will be briefly discussed.

The purpose of this report is to define the problems associated with aircraft fuel instability, review what is currently known about the problem, describe the research program sponsored by NASA Lewis, and identify those areas where more research is needed. The term fuel instability generally refers to the gums, sediments, or deposits which can form as a result of a set of complex chemical reactions when a fuel is stored for a long period at ambient conditions or when the fuel is thermally stressed inside the fuel system of an aircraft.

Thermal instability was first identified as a problem in aviation turbine engines in the 1950's. During the 1960's, early studies in the United States on the supersonic transport (SST) gave considerable attention to the problem of fuel instability, because in this SST, the fuel was to be used as a heat sink for the wing surfaces which are heated aerodynamically. It is generally acknowledged that current aircraft turbine fuels do not present a significant problem with regard to fuel instability for current subsonic aircraft. However, turbine fuels with broadened properties or nonpetroleum-derived fuels (from shale, tar-sands, coal, etc.) may have reduced thermal stability because of their higher content of olefins, heteroatoms, and trace metals. Moreover, advanced turbine engines may increase the thermal stress on fuels because of their higher pressure ratios and combustion temperatures. Deposition of solids within the fuel system of aircraft may lead to fouled heat exchangers, plugged fuel nozzles, and/or jammed fuel valves which may result in excessive oil temperatures and non-uniform fuel spray patterns which could cause distorted turbine inlet temperatures (hot spots).

During the past several years NASA Lewis has been engaged in a research and technology program to determine the effects of broadened-property fuels on engine and fuel system components and to evolve the technology needed to use these fuels. Broadening fuel properties may offer the potential for increasing the refinery yield of jet fuel. Moreover, additional energy intensive treatment of poorer quality crudes and syncrudes will be required if jet fuel with current properties is to be produced. One of the major problem areas that must be addressed is fuel instability because of the reasons given in the previous paragraph. In recognition of its importance, NASA Lewis has established a broadly-based research program to better understand the underlying causes of fuel thermal degradation. Our in-house research is supported by grants with universities and contracts with industry. The progress, status, and results for these various activities will be reviewed and discussed in the report, along with some preliminary thoughts on design approaches required to minimize the effects of lowered thermal stability. In order to place the NASA Lewis program on fuel instability in its proper perspective, we should point out that similar research programs are currently being sponsored by other U.S. Government agencies, particularly the Department of Defense. These programs have made significant contributions toward the understanding of fuel degradation processes and effects. However, it is beyond the scope of this report to discuss these programs in any detail, and they will only be mentioned in the course of providing background for the NASA-sponsored program.
The thermal decomposition of hydrocarbons can be approximately classified into three temperature regimes as follows:

1. **Low temperature** - below 300°C: decomposition by autoxidation
2. **Intermediate temperature** - between 300° and 500°C: decomposition by further reactions of autoxidation products and by some direct pyrolysis of the fuel molecules. High temperature - above 500°C: decomposition by direct pyrolysis of the fuel

The first two of these temperature regimes are operative in the fuel system of aircraft. It is desirable to understand the chemical mechanism or mechanisms involved in fuel degradation in order to facilitate coping with the problem of thermal instability. Because deposit formation is such a complex process, this can best be done by first obtaining both deposit formation and chemical kinetics information for less complicated systems, namely pure hydrocarbons and binary mixtures. This knowledge can then be applied to the more complex mixtures in practical fuels. One of the first studies of this kind was reported in 1977 by Hazlett at the Naval Research Laboratory. He used a jet fuel thermal oxidation tester (JFTOT) connected to a gas chromatograph to study the thermal decomposition of pure n-dodecane in the presence of dissolved oxygen. (The JFTOT will be described in a later section of this report.) He measured dissolved 02 consumption and formation of the stable intermediate hydroperoxide ROOH as a function of temperature. The mechanisms he proposed for the reactions occurring in the low and intermediate temperature regimes are presented in Fig. 1. Reaction 1 indicates the abstraction of a hydrogen atom from the fuel RH by a reactive molecule or free radical X to form the reactive alkyl radical R-which then reacts (reaction 2) with O2 to form the unstable ROOH radical. ROOH reacts mainly by reaction (3) with the fuel to form the stable hydroperoxide ROOH and regenerate the alkyl radical. At much smaller extent ROO- undergoes unknown reactions to form ultimately the harmful solid deposit and sediment. Hazlett and other investigators have shown that the amount of deposit formation is not directly related to the rate of primary fuel oxidation at low temperatures, even though the oxidation must take place in order for any solid formation to occur. Reactions of intermediate oxidation products lead to solid formation, and these reactions are not understood at present.

At intermediate temperatures the secondary reactions involve the decomposition of ROOH (reaction (5)) to form ketones, alcohols and aldehydes (reactions (6) - (8)). In addition, direct pyrolysis of the fuel molecule, also a complicated chemical process, starts to occur as shown in reaction (9). Isomerization and decomposition of the radicals formed by pyrolysis leads to olefin formation. It is clear from the complexity of this partial mechanism for a pure hydrocarbon's decomposition that the elucidation of the degradation mechanism for a practical fuel will be a formidable task indeed.

**Deposit Characteristics**

The fuel deposits that form in aircraft fuel systems may occur as soft gums, as strongly adhering lacquers and varnishes, or as brittle cokes. Studies of the morphologies of these deposits [6,7] indicate that a generally an agglomeration of microspheres, although some plate and rod forms have also been observed. Chemical analysis of fuel deposits has revealed these additional general characteristics:

1. The hydrogen/carbon ratio is lower in the deposits than in the original fuel,
2. Oxygen concentration of the deposits is much greater than in the thermally unstressed fuel, and
3. Other heteroatoms such as nitrogen and sulfur are highly concentrated in the deposits, with concentrations several orders of magnitude higher than in the fuel.

The high concentration of heteroatoms in the deposits relative to their concentrations in the fuel is strongly supportive of the importance of these trace organic impurities in the deposit formation process. The lower hydrogen/carbon ratio in the deposits suggests that aromatic compounds play an important role in deposit formation.

**EVALUATION OF INSTABILITY**

Attempts to measure the amount of deposit produced by a fuel under a given set of conditions have ranged from small-scale glass laboratory devices to full-scale fuel system simulators. These devices have been thoroughly reviewed in ref. 2, and this report will concentrate on the American Society for Testing and Materials (ASTM) test methods for evaluating fuel thermal stability and on the larger dynamic fuel stressing rigs commonly referred to as laboratory simulators.

**ASTM Test Methods**

The first standard specification test for fuel thermal stability was the CRC Fuel Coker Test (ASTM D-1860) which was designed to simulate the fuel manifold conditions that had led in the 1950's to the first operation problems caused by fuel instability [8]. The test device consists basically of a heated tube around which fuel flows at a rate of 4.5 liters/hr, followed by a heated 25 micron filter of sintered stainless steel. The former simulates the engine's hot fuel lines, and the latter the fuel nozzles and passages where fuel deposits and sediments could be trapped. The heated tube is maintained at 150°C and the filter at 204°C, with a system pressure of 170 kPa (150 psig). Fuel is flowed through for 5 hours. A fuel is rated on the basis of color change on the heated tube and on pressure drop across the filter.

Because of some difficulties with the CRC Coker, primarily large sample size, low operating pressure, and long test duration, this device has been replaced by the JFTOT (ASTM D-3241), which is now used to determine if an aviation turbine fuel meets the current thermal stability requirement. A schematic of
the JFTOT is shown in Fig. 2. A small aluminum tube is heated electrically to the desired test temperature, and fuel from the reservoir is pumped through an annulus surrounding the tube, out through a test filter, and then back to the upper portion of the fuel reservoir. If the fuel is unstable, deposits will form on the heated tube, and any particulates formed will lead to an increase in pressure drop across the filter. The standard JFTOT procedure consists of flowing 0.18 liters/hr of an aerated fuel at 3447 kPa (500 psig) nitrogen gas pressure over the heated tube (260° C) for 2 1/2 hours. The amount of deposit on the tube can be rated either visually, on a scale of 0 (no deposit) to 4 (heavy deposit), or by using a tube deposit rating (TDR) that operates on a light reflectance principle. The visual rating is the method currently called for by ASTM D-3241, and a value of 3 or less is required to meet the current specification for thermal stability. However, the TDR is frequently used in fuels research, and its scale ranges from 0 to 50, with a value of 12 to 15 generally used as an equivalent criterion for passing the standard JFTOT test. In addition to the requirement on deposits, the A value across the filter must not exceed 3.3 kPa (25 torr) by the end of the 2 1/2-hour test if the fuel is to pass the test. By operating the JFTOT at temperatures other than 260° C, the temperature at which a fuel just fails either of the above tests may be determined. This is referred to as the "breakpoint temperature" and is used to compare the thermal stability of fuels.

Tests for storage stability usually involve placing the fuel in a small container and aging it at temperatures sufficient to accelerate the instability process, usually 80° to 100° C, so that measurable deposits are produced in a reasonable time. An ASTM procedure (D973) has been used to predict storage stability. Fuel was placed in an oxygen bomb for 16 hours at 100° C. However, experience showed that any fuel which passed the JFTOT test (breakpoint less than 260° C) always passed this storage test. Because of this, the specification for storage stability has been dropped [3].

Laboratory Simulators

There is a large gap between the operating conditions of a device such as a JFTOT and the conditions to which fuel is subjected in an actual aircraft fuel system. A laboratory simulator is an apparatus designed to bridge this gap. Important parameters such as inlet temperature, wall temperature, fuel flow rate, residence time, Reynolds number, pressure, etc. can be accurately controlled and are representative of actual engine conditions. A schematic of a laboratory simulator used in recent thermal stability studies at the United Technologies Research Center (UTRC) under NASA sponsorship [9] is shown in Fig. 3. Two closely fitted, titanium alloy blocks formed the flow channel which had a cross-section 0.254 cm (0.10 in.) high by 2.84 cm (1.12 in.) wide. Because of the high thermal conductivity of this alloy, the wall temperature of the channel was very uniform. All surfaces were plated with nickel and then gold to minimize material effects. Four specimen mounts were attached along the length of the test section. Each held a stainless-steel disc that was used to determine deposit formation rate by weighing the test disc on a microbalance before and after the experiment. Heating was accomplished by cartridge heaters in both the top and bottom plates. Pressure atomizing nozzles were used to control the flow rate and to observe any flow instabilities in the test section. Test temperatures ranged from 149° to 399° C (300° to 750° F), test pressures from 690 to 2070 kPa (100 to 300 psig), and fuel flow rates from 2.14 to 21.4 grams/sec.

A laboratory simulator of somewhat similar design has been built at NASA Lewis for in-house research and is currently being checked out. The entire test section and flow system were constructed of stainless steel to avoid any possibility of accelerated fuel degradation from copper materials. The NASA facility also has space for four test specimens for each experimental run, but it is capable of a wider range of operating conditions than the UTRC simulator. The flat specimens provided by both the NASA and UTRC simulators are significantly easier to characterize by most current analytical techniques than the tubular deposits produced by the JFTOT and Fuel Coker. Because of this advantage for a flat test specimen, NASA Lewis has modified one of its JFTOTs for use with a flat sample holder [10].

DESCRIPTION OF RESEARCH

Research on fuel instability can be classified into two general types: research designed to understand chemical mechanisms and research on the behavior of actual fuels. When the elucidation of chemical mechanisms is the main objective of the research, pure compounds or "model fuels" made up of a mixture of a few pure compounds are usually employed in order to simplify the chemistry. Compounds containing heteroatoms or trace metals are then added in small amounts to determine their effects on fuel stability. Other research efforts concentrate on the deposits produced by actual fuels when they are thermally stressed under controlled conditions. As has previously been discussed in this report, practical fuels are so complex and contain such a variety of trace impurities that mechanism studies using them would be extremely difficult. Common to both these types of research is the need to characterize the resulting fuel deposits. A variety of analytical techniques is required to provide information on both elemental and molecular composition. NASA Lewis' current work on fuel stability, both in-house and under grant or contract, will be reviewed according to these three categories of research.

Chemical Mechanism Research

A modified JFTOT procedure was used in an in-house study to evaluate deposit and sediment formation for four aerated pure hydrocarbon fuels over the tube temperature range 150° to 450° C [11]. Stainless steel heated tubes rather than the usual aluminum tubes were used to permit higher test temperatures and to increase the deposition rate and thereby reduce test time. Four fuel types were studied: a normal alkane (n-decane), a naphthenic (cyclohexane), and an aromatic (benzene). Some results from this study are shown in Fig. 4, where tube deposit amounts, expressed as net TDR values, are plotted against tube position. TDR values are approximately proportional to deposition rate. At the lower temperatures of 150° and 250° C, all four of the fuels produced only small amounts of deposit during the 40 minute run, with TDR values less than 5. At 350° C the olefinic fuel, 1-hexene, formed the largest amount of deposit, cyclohexane the second largest, with the other two fuels forming much less deposit. The fact that 1-hexene formed the most deposit is consistent with its susceptibility to oxidation and particulate formation at higher temperatures of 400° and 450° C all fuels except benzene formed large amounts of deposit, with TDR values of 40 to over 50. It is not surprising that benzene formed much less
deposit that the other three fuels. This can be attributed to the benzene aromatic structure, which is highly resistant to oxidation [12]. However, substituted benzenes and multi-ringed compounds are generally less resistant to oxidation. Sediment formation was also measured in these experiments and is shown in Fig. 4 as a black line. For these fuels only benzene shows any significant increase. Thus in these experiments, the aromatic benzene appears to form considerable sediment but very little deposit.

This research effort utilizing a modified JFTOT to thermally stress hydrocarbons is continuing. Effects of deoxygenation and pre-stressing of the fuel are currently under investigation. Liquid and gas chromatography as well as mass spectrometry will be used to determine reaction products, including chemical precursors that ultimately lead to deposits, in an effort to increase our understanding of the decomposition mechanisms involved in the fuel degradation process.

Effects of Fuel Nitrogen - We have already mentioned that heteroatoms such as nitrogen that are present in fuels seem to be concentrated in the fuel deposit. A number of experiments have been carried out in which a fuel is spiked with small amounts of nitrogen-containing compounds that are known to occur in actual fuels, in an effort to determine their effect on fuel deposit rates. Some results of this type experiment based on data from ref. 13 are shown in Fig. 5(a). Deposit rate for Jet-A and Jet-A spiked with 0.1 percent N from 2,5 dimethylpyrrole is plotted against temperature. It is clear from the figure that the deposit rate for the spiked Jet-A is dramatically higher at a given temperature, and that the increase in deposit rate with temperature is also considerably greater. In Fig. 5(b) the breakpoint temperature for Jet-A and Jet-A spiked with 0.01 to 0.1 percent nitrogen from pyrrole (a single-ringed N-containing compound) and indole (a double-ringed N-containing compound) are compared [14]. Since breakpoint depends on the rate of deposit formation, based on the results shown in Fig. 5(a), one would expect a reduced breakpoint for the spiked fuels. This is indeed what was observed. The Jet-A spiked with pyrrole and indole have breakpoint temperatures about 40°C and about 20°C less than the unspiked Jet-A, respectively.

Under a NASA grant with the Colorado School of Mines, Daniel has also studied the effect of various organic nitrogen compounds on deposit formation in Jet-A and Jet-A-

- ESR Studies of Fuel Instability - An electron spin resonance (ESR) spectrometer responds to the presence of an unpaired electron in a chemical species. Such species are by definition free radicals, and we have already pointed out that the chemistry of fuel deposit formation involves primarily free radical reactions. Thus, an ESR spectrometer capable of operating at high temperatures and pressures can be used to directly monitor some of the important reactions occurring during deposit formation. We believe that the use of ESR to study fuel degradation offers the potential for significantly increasing our understanding of the reaction mechanisms and kinetics involved in this complex process. Accordingly, NASA Lewis has recently entered into an Interagency Agreement with the Oak Ridge National Laboratory (ORNL) to study free radical reactions in model compounds representative of constituents of jet fuel using ESR.

In previous research at ORNL involving coal liquids, Livingston and Zeldes developed a flow apparatus to use with an ESR spectrometer to study fluids at high temperatures and pressures [18]. Only minor modifications to this apparatus were required to enable work on fuel thermal stability to be undertaken. The
flow system used in conjunction with the ESR spectrometer is shown in Fig. 8. A liquid sample contained in a reservoir was pumped through a capillary of the ESR spectrometer to remove dissolved oxygen. The liquid flows to a positive displacement high pressure pump, and pressures from 70 to 135 atm were used with a flow rate of 1 ml/min. There is provision for injecting a gas such as oxygen at high pressure. The liquid flows through a capillary that traverses the microwave cavity of the ESR spectrometer and then to a back pressure regulator where the system pressure is set. The liquid exiting the regulator is at atmospheric pressure and may be recirculated as shown in Fig. 8 or collected without recirculation for subsequent analysis. Thermal production of radicals is achieved by heating the capillary with air that has been heated electrically. Photolysis is performed by a vacuum jacketed silica tube, and the air flows in the annular space. Photolysis is also performed photolytically. This is done by focusing UV from a 500 W high pressure mercury arc onto the sample within the capillary. Photolysis is used at room temperature and up to temperatures where thermal decomposition becomes fast enough to see signals. Typical steady-state concentrations of radicals from hydrocarbons are in the micromolar region with mean lifetimes ranging downwards from typical room temperature values near 1 millisecond.

A wide variety of model compounds is being examined that give observable spectra at temperatures up through the intermediate regime (through 300 °C). Compounds include n-dodecane, aromatic hydrocarbons, unsaturated hydrocarbons, and sulfoxides. Much of the work centers upon oxidizing effects and includes the use of peroxides and hydroperoxides as free radical initiators. The initial phase is a broadly based exploratory survey to search out major reaction pathways that will subsequently be examined in detail.

Deposit Formation in Practical Fuels

The deposit formation rate of actual fuels is usually determined in a flow device generally referred to as a laboratory simulator. Such devices are also capable of providing relatively heavy deposits on flat test specimens which are more easily analyzed and characterized than JFTOT tubes. The schematic of a laboratory simulator used by Vrmos under contract to NASA Lewis was described previously in the discussion of Fig. 3. Vrmos has fixed this apparatus to determine the thermal stability of four actual hydrocarbon fuels under conditions representing operation of an aircraft gas turbine engine [9, 19]. Surface temperature was varied from 149 °C to 399 °C (300 °F to 750 °F) for fuel flows of 0.64 to 12.8 grams per second (0.75 to 15 gal/hr) at pressures of 2040 kPa (300 psig). Test times varied from 1 to 8 hours.

The rate of deposition was obtained by measurement of the weight gained by metal discs located along the channel wall.

Figure 9 shows representative results for the four fuels using stainless steel test discs: Jet A, home heating oil, a straight-run diesel which has no cracked stock, and a normal diesel which has a large cracked fraction. Below 260 °C (500 °F), where there is an apparent change in mechanism, Jet A, straight-run diesel, and the cracked diesel all had comparable deposition rates. Above 260 °C, Jet A had the lowest rate, straight-run diesel somewhat higher, and cracked diesel the highest. Up to 260 °C the home heating oil had the highest rate while above 260 °C it was lower than the cracked diesel.

The effect of wall material was determined by using stainless steel, aluminum, and brass test discs. Stainless steel and aluminum gave comparable results while the brass specimens gave a significant increase in deposition rate. Tests were also conducted in a static reactor at temperatures of 149 °C to 427 °C (300 °F to 800 °F) for times of 1 minute to 24 hours. Much smaller deposition was found, indicating the importance of fluid transport in the deposition mechanism. As-received and stressed fuels were examined by liquid chromatography in an attempt to detect any changes in fuel composition caused by the heating. It is known that the changes being sought are very small and therefore were found, probably due to the limits of detection of the instrument.

The strong temperature dependence of deposit rate suggests that an Arrhenius plot, log of deposition rate versus reciprocal temperature, might correlate the data. Figure 10 is such a plot by Vrmos [9] of data for Jet-A at two different flow rates. The overall activation energy, given by the slope of the lines, is in reasonable agreement with some previous results for Jet-A [20]. At higher flow rates, the deposit rate is lower at the same wall temperature, yet the temperature dependence, or curvature, is approximately the same. The lower deposit rate at increased flow points out the importance of bulk-fluid heating and homogeneous reaction effects on the formation of deposits. In both cases, the magnitude of the activation energy, which is much less than that for typical homogeneous reactions, suggests a surface-catalyzed process.

Additional deposit-rate studies are planned for the NASA laboratory simulator previously described. In addition, this simulator along with the NASA JFTOT modified for flat test samples, will be used to generate deposits for detailed chemical and physical analysis.

Deposit Analysis

Due to the complex and intractable nature of the deposits, as well as the small quantities generally available for analysis, the detailed molecular structure of the deposits has remained largely unknown. Recent attempts to perform detailed characterization of the deposits have involved the use of a variety of modern analytical instrumental techniques which are listed in Table I along with a brief comment about the type of information each technique is capable of producing. Each technique has the potential to provide a portion of the structural information sought, but each has specific limitations as well. Through in-house research efforts and sponsored studies, each of the techniques listed is or will soon be employed to characterize deposits obtained from the JFTOT flat sample holder and the NASA thermal stability laboratory simulator under varied but carefully-controlled conditions. Both well-characterized commercial fuels and model fuels prepared from representative pure compounds are being used. To provide additional information, a deposition mechanism, an identification of the deposit precursors, and the sensitivity of the formation rates to fuel stressing conditions. These results
should ultimately lead to deposit prevention through the knowledgeable development of fuel additives, the types and levels of refinery fuel treatment required, and future fuel system designs.

DESIGN CONSIDERATIONS

We stated in the introduction that in the future, the thermal stability of fuels may be lower and the thermal stresses to which they are subjected, higher. One way to alleviate this problem is by additional processing at the refinery. Another is for the airlines to decrease the operating time between inspections or component removal and maintenance. Either of these approaches might be prohibitive in terms of increased costs to the airlines.

The most cost effective approach to utilizing fuels of reduced thermal stability in new, higher efficiency engines is to incorporate, in the design stage, components and designs tolerant to variations in this fuel property. It is beyond the scope of this report to discuss in any detail the types of design changes that might be required, but a few comments are appropriate.

In contracted analytical studies with NASA Lewis, General Electric and LockheedCalifornia have been analyzing and assessing advanced fuel system concepts. Both contractors have identified thermal stability and freezing point as the two most important fuel properties affecting fuel system performance. They both have studied advanced fuel systems capable of using a fuel with a breakpoint temperature of about 205°C, considerably below the current specification for thermal stability (breakpoint of at least 260°C). Although the final results for these two studies are not yet available, the advanced concepts under consideration include the following approaches to reducing heat load on the fuel after it leaves the fuel tank: (1) relocate valves to lower temperature environment and use low p nozzles which permit larger injector ports, (2) use variable displacement pumps instead of the current gear type to reduce pump heat rejection to the fuel, (3) reject lubricating-oil heat to the bulk of the fuel in the fuel tanks rather than to the much smaller amount of fuel metered for the combustor, and (4) improve the thermal tolerance of fuel nozzles and fuel lines exposed to high heat load by using heat shields, insulation, and/or cooling air. While concepts such as these may potentially allow the use of fuels with reduced thermal stability, none of these concepts has been verified experimentally. Some additional words of caution are in order. At a workshop held at NASA Lewis [21], it was pointed out that heat shield and insulation devices designed to reduce heat soak-back to fuel nozzles might reduce the aerodynamic efficiency of air-blast nozzles. This would adversely affect the atomization quality of the fuel spray which is already an area of concern with alternative fuels. This example indicates that we must be wary of interaction effects when changing the design of fuel systems.

CONCLUDING REMARKS

Although fuel instability is not a serious problem for current fuels and aircraft, it could well be a major problem in the future because of the possible deterioration of the quality of petroleum crudes and the use of syncrudes. The problem could be exacerbated by the increased thermal stress that will probably be encountered in advanced turbine engines. We have tried to show in this paper that the instability of aircraft fuel is an extremely complex problem, with experimental investigations yielding results that sometimes appear to be contradictory. Although a great deal of work has been done to understand the mechanisms of deposit formation, deposit composition, the role of trace impurities, and the effects of fuel instability on airframe and engine fuel systems, much work remains to be done. It is anticipated that an increased understanding of fuel instability will lead to methods for minimizing deposit formation or of ameliorating its effects on fuel systems.

There is another potential benefit from an increased understanding of the fuel degradation process which is related to fuel processing. Petroleum refineries currently do not use any specific processing treatment to control the thermal stability of jet fuel, although selective chemical additives (anti-oxidants) are frequently used to improve storage stability. The desired level of thermal stability is a by-product of the other processing the fuel must undergo in order to meet other specifications. This practice will probably not be satisfactory for alternative fuels because of reasons already stated. However, a fundamental knowledge and understanding of the fuel degradation process will probably provide value insight as to what type or types of additional processing should be undertaken to improve the thermal stability of the resulting fuel.

Although there are many facets of the fuel instability problem that need to be explored, it is the authors’ opinion that additional research with the following objectives will lead to significant benefits in understanding and dealing with fuel instability:

- Establish the roles and degree of importance of fuel hydrocarbon type and trace contaminants in determining fuel instability.
- Compare rates, morphologies, and compositions of deposits produced via simulators with actual deposits produced in an engine fuel system.
- Evolve a model which will predict fuel deposition rates in an actual engine fuel system based on input from laboratory simulators.
- Establish the relationship between storage deposits and thermal deposits. Are the mechanisms the same, precursors the same?
- Determine the effectiveness of additives (anti-oxidants, free-radical scavengers, dispersants) for improving fuel stability as well as their interactions with each other and fuel components.
- Experimentally evaluate conceptual fuel system components and fuel injectors that reduce exposure to thermal stress and are more tolerant to small amounts of deposition.
REFERENCES


### TABLE I. - MODERN INSTRUMENTATION FOR DEPOSIT CHARACTERIZATION

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LOW TEMPERATURE
(BELOW 300°C - AUTOXIDATION)

1. \( \text{RH} + X \rightarrow \text{R} \cdot + \text{HX}, \text{(INITIATION)} \)  
2. \( \text{R} \cdot + \text{O}_2 \rightarrow \text{RO}_2 \cdot \)  
3. \( \text{RO}_2 \cdot + \text{RH} \rightarrow \text{ROOH} + \text{R} \cdot \)  
4. \( \text{RO}_2 \cdot + \text{?} \rightarrow \text{DEPOSITS, GUMS, SEDIMENTS} \)

INTERMEDIATE TEMPERATURE
(300 - 500°C)

5. \( \text{ROOH} \rightarrow \text{RO} \cdot + \text{HO} \cdot \)  
6. \( \text{RO} \cdot + \text{RH} \rightarrow \text{ROH} + \text{R} \cdot \)  
7. \( \text{HO} \cdot + \text{RH} \rightarrow \text{H}_2\text{O} + \text{R} \cdot \)  
8. \( \text{ROOH} + X \rightarrow \text{KETONE} + \text{XH}, \text{(SECONDARY HYDROPEROXIDE)} \)  
9. \( \text{RH} \rightarrow \text{R}_1 \cdot + \text{R}_2 \cdot, \text{(PYROLYSIS)} \)

\( \text{ISOMERIZATION AND DECOMPOSITION OF R}_1 \cdot \)  
\( \text{AND R}_2 \cdot \text{TO FORM OLEFINS} \)

Figure 1. - Proposed mechanisms for fuel degradation.
Figure 2. - Schematic of jet fuel thermal oxidation tester (JFTOT).
Figure 4. Tube deposit and sediment formation for four hydrocarbon fuels.
Figure 5. - Effect of fuel nitrogen content on thermal stability.

(a) Effect of fuel nitrogen content on deposit formation rate.

(b) Breakpoint temperatures of Jet-A and solutions of nitrogen compounds in Jet-A ranging from 0.01 to 0.1 weight percent N.
Figure 7 - Oxidation of n-dodecane with indene at 130°C. (Gum is in mg/100 g fuel, determined at 105°C.)

Figure 6 - Effect of added nitrogen concentration on deposit rate at 130°C.
Figure 8. - Flow system for fuel thermal stability studies using electron spin resonance (ESR) spectrometer.

Figure 9. - Deposition rate of various fuels.
Figure 10. - Arrhenius plot of deposit rate in Jet A.

FLOW RATE
- 2.14 x g/sec (2.5 gal/hr)
- 6.85 x g/sec (8 gal/hr)

RATE = \( A \exp \left( \frac{-E}{RT} \right) \)

- 9 kcal/mol
- 10 kcal/mol

(SURFACE TEMPERATURE, 10^3/K)
Current aircraft turbine fuels do not present a significant problem with fuel thermal stability. However, turbine fuels with broadened properties or nonpetroleum-derived fuels may have reduced thermal stability because of their higher content of olefins, heteroatoms, and trace metals. Moreover, advanced turbine engines will increase the thermal stress on fuels because of their higher pressure ratios and combustion temperature. In recognition of the importance of this problem, NASA Lewis is currently engaged in a broadly-based research effort to better understand the underlying causes of fuel thermal degradation. The progress and status of our various activities in this area will be discussed in this paper. Topics to be covered include: nature of fuel instability and its temperature dependence, methods of measuring the instability, chemical mechanisms involved in deposit formation, and instrumental methods for characterizing fuel deposits. Finally, some preliminary thoughts on design approaches for minimizing the effects of lowered thermal stability will be briefly discussed.