TURBINE HEAT TRANSFER

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Improved turbine durability and performance and reduced development cost will all result from improved methods of predicting turbine metal temperatures. As you know, better metal temperature prediction methods require improvements in the method of determining the hot gas flow through the turbine passage and the cooling air flow inside the airfoil and in the methods of predicting the heat transfer rates on the hot gas-side and on the coolant-side of the airfoil. The overall turbine heat transfer effort is directed at improving all four of these areas of concern.

Achievement of these improvements requires a rigorous and systematic research effort from both the experimental and analytical sides. The experimental approach being pursued starts with fundamental experiments with simple shapes and flat plates; progresses on to more realistic cold, warm, or hot cascades; continues to progress on to more realistic warm turbine, large low-speed turbine, or transient turbine tests; and finally combines all the interactive effects in real-engine environment turbine tests. Analytical approaches being pursued also start with relatively simple mathematical models and progress to more realistic cases that include more interactive effects, and finally combines all the interactive effects of the turbine operating in the real engine environment.

Currently, contract and grant activities are being (or will be) conducted to obtain fundamental experimental data and to develop and/or compare analytical methods in all four areas of concern. These contract and grant activities will be discussed in detail later in this meeting by the respective principal investigators.

Major NASA Lewis in-house turbine research efforts are being pursued to obtain more realistic and real-engine type turbine experiments. The NASA
Lewis Research Center is in the process of activating our High Pressure-High Temperature Facility (HPF) with initial 20 atmosphere and 2500°F experimentation scheduled for the last quarter of 1982. HPF will provide the country with a known real-engine environment in which to conduct controlled aerothermodynamic and structural research studies. We envision a multiple role for HPF in providing engineering-quality research data for modeling and code verification, in defining a real-engine environment, and in evaluating advanced turbine cooling technology in a real-engine environment.

The major turbine research parameters of interest that will be measured or determined to provide a better understanding of the thermal, aerodynamic, and mechanical performance of air-cooled turbine airfoils are the following:

1) local hot gas recovery temperatures along the airfoil surfaces,
2) local airfoil wall temperature,
3) local hot gas-side heat transfer coefficients on the airfoil surfaces,
4) local coolant-side heat transfer coefficients inside the airfoils,
5) local hot gas flow velocities and secondary flows at real-engine conditions, and
6) local delta strain range of the airfoil walls.

Currently, little of this type experimental research information exists with controlled warm or real-engine conditions and known boundary conditions.

These in-house turbine research efforts will be conducted using the best available analyses to help define the test configurations, the types of research measurements, and/or the test conditions and for the comparison with the measured research results. Analytical efforts will initially use the best available flow and heat transfer codes such as a two- or three-dimensional inviscid flow code and a two- or three-dimensional boundary layer heat
transfer code. These analyses will be applied at the mid-span section and possibly at the hub and tip sections or other local zones of the passage. More sophisticated three-dimensional viscous codes and three-dimensional viscous codes with boundary layer resolution will be used as they become available. These analytical efforts will be conducted using the best available source or sources in-house and on contract with industry and universities.
OBJECTIVES: IMPROVE ACCURACY OF PREDICTING LOCAL BLADE METAL TEMPERATURES USING COMPUTER CODES THAT ARE COMPATIBLE WITH STRUCTURAL ANALYSIS CODES

APPROACH:

- INVESTIGATE HOT GAS STREAM AND COOLANT PASSAGE FLOW MECHANICS AND HEAT TRANSFER
- OBTAIN BENCHMARK-QUALITY AND ENGINEERING-QUALITY DATA FOR EVALUATION AND IMPROVEMENT OF MODELS PRESENTLY USED IN PREDICTION CODES
- UTILIZE IMPROVED MODELS TO IMPROVE ACCURACY OF PREDICTING LOCAL GAS-SIDE AND COOLANT-SIDE HEAT TRANSFER COEFFICIENTS
- UTILIZE FLOW MODELS TO IMPROVE ACCURACY OF PREDICTING LOCAL HOT GAS STREAM ENVIRONMENT THROUGH TURBINE ROWS AND COOLANT FLOW CONDITIONS INSIDE THE AIRFOIL
- INTEGRATE IMPROVEMENTS IN PREDICTION OF HOT GAS STREAM ENVIRONMENT AND COOLANT FLOW AND LOCAL HOT GAS-SIDE AND COOLANT-SIDE HEAT TRANSFER COEFFICIENTS INTO IMPROVED METAL TEMPERATURE PREDICTION CODES
- PROVIDE ENGINEERING-QUALITY TEST CASES FOR EVALUATION OF ACCURACY OF PREDICTION AND INPUT TO STRUCTURAL ANALYSIS CODES
TURBINE ENGINE HOT SECTION TECHNOLOGY

BUILDING BLOCK AEROTHERMAL TURBINE RESEARCH APPROACH

REAL WORLD
NASA HIGH PRESSURE TURBINE

HIGH SPEED WARM TURBINE
TRANSIENT TURBINE
LARGE LOW SPEED TURBINE

ROTATION/UNSTEADY

AIRFOIL SPECIFIC
STATIC, CASCADES
COLD ANNULAR AND WARM ANNULAR

WIND TUNNELS AND FUNDAMENTAL MODELS
TRANSITION, VISUALIZATION, TURBULENCE, FILM COOLING

EXPERIMENTAL APPROACH
ANALYTICAL APPROACH
CODE DEVELOPMENT
ANALYTIC MODELING
CODE VERIFICATION
## TURBINE HEAT TRANSFER
### PLANNING SCHEDULE

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<tr>
<th>ELEMENT</th>
<th>FY</th>
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<th>EXPECTED RESULTS</th>
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<td>GAS-SIDE HEAT TRANSFER, NON-ROTATING, FILM</td>
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<td>--OBTAIN BENCHMARK QUALITY AEROTHERMODYNAMIC DATA AND IMPROVED THREE-DIMENSIONAL VISCOUS FLOW CODES, NO ROTATION</td>
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<td>GAS FLOW ENVIRONMENT AND HEAT TRANSFER, ROTATING, 2-D</td>
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<td>--IMPROVED HEAT TRANSFER CORRELATION AND MODEL FOR IMPINGEMENT COOLING</td>
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<td>COOLANT SIDE HEAT TRANSFER WITH ROTATION AND ENTRANCE GEOMETRY</td>
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<td>--HEAT TRANSFER CORRELATIONS, INCLUDING EFFECTS OF ROTATIONS AND ENTRANCE GEOMETRY</td>
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<td>METAL TEMPERATURE PREDICTION CODES</td>
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<td>--METAL TEMPERATURE PREDICTION CODES WITH IMPROVED HEAT TRANSFER MODELS/CORRELATIONS</td>
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<td>IN-HOUSE RESEARCH AND VERIFICATIONS</td>
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<td>--VERIFICATIONS OF FLOW, HEAT TRANSFER METAL TEMPERATURES, AND STRAIN PREDICTIONS, AT NEAR AND REAL-ENGINE TYPE CONDITIONS</td>
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<td>Possibly Contract and/or In-House:</td>
<td>Metal Temperature Prediction Codes</td>
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<td><strong>Scope:</strong></td>
<td>Review and modify existing airfoil metal temperature codes for efficient incorporation of developed models and for interfacing with structural analysis codes</td>
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<td>• Incorporate improved models developed under HOST</td>
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<td>• Assure efficient interfacing with structural codes</td>
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TURBINE HEAT TRANSFER

IN-HOUSE RESEARCH AND VERIFICATIONS

SCOPE: Experiments and analysis to supplement contractual and grant efforts on improving accuracy of flow environment and heat transfer predictions and the verification of developed/improved prediction methods.

DURATION: Six years

APPROACH:
- Measure local heat transfer coefficients over a stator vane at near-real engine conditions and compare with prediction.
- Procure laser anemometer system and install and check-out in NASA warm turbine.
- Obtain measurements in NASA high pressure turbine (HPT) at near real engine conditions to evaluate prediction accuracies of codes for hot gas flow, hot gas environment, hot gas-side and coolant-side heat transfer coefficients, metal temperatures, and strain.
TYPICAL AIRFOIL AEROTHERMODYNAMIC ANALYSIS METHOD

ENGINEERING MEASUREMENTS TO DEFINE

HOT GAS BOUNDARY CONDITIONS

TWO- OR THREE-DIMENSIONAL, NON-VISCOS OR VISCOS FLOW CODE

BOUNDARY LAYER HEAT TRANSFER CODE

ENGINEERING MEASUREMENTS TO DEFINE AND VERIFY CODE

ENGINEERING MEASUREMENTS TO VERIFY CODE

ENGINEERING MEASUREMENTS TO VERIFY

NODE GENERATOR

THREE-DIMENSIONAL STEADY STATE AND TRANSIENT CONDUCTION CODE

ENGINEERING MEASUREMENTS TO MEASURE AND VERIFY

WALL TEMP. AND PRESS.

STRUCTURAL ANALYSIS CODE

LIFE PREDICTION CODE

COOLANT-SIDE HEAT TRANSFER CORRELATIONS

ONE-, TWO-, OR THREE-DIMENSIONAL VISCOS FLOW CODE OR NETWORK CODE

POSSIBLY ENGINEERING MEASUREMENTS TO MEASURE AND VERIFY
ENGINEERING-TYPE MEASUREMENTS DESIRED

- $T_{g,4}, T_{g,4}$
- $V_{g,4}$
- $P_{g,4} \rightarrow V_{g,s}$
- $q/A_{tot}$
- $q/A_{rad}$ on endwall

- $T_{g,5}, T'_{g,5}$
- $V_{g,5}$
- $V_g, V'_g$
- $q/A_{tot}$ on tip shroud

- $W_c, T_c$
- $P_{g,6}$

- $q/A_{rad}$ on endwall
NONINTRUSIVE VELOCITY MEASUREMENTS THROUGH COMPLETE TURBINE STAGE WITH UNIFORM AND NONUNIFORM INLET TEMPERATURE PROFILES

ANNULAR CASCADE AT 50% AXIAL CHORD

SUCTION SURFACE

90.3% SPAN

PRESSURE SURFACE

BENCHMARK DATA FOR EVALUATION OF ADVANCED 3-D FLOW CODES FOR AERODYNAMICS AND HEAT TRANSFER
HIGH PRESSURE AND TEMPERATURE TURBINE RESEARCH FACILITY: REAL ENGINE ENVIRONMENT

- SINGLE STAGE 20" TIP DIA
- TIT 2000 TO 4000°F
- TIP 600 psia
- WHEEL SPEED LIMIT 23,000 rpm
- AIR FLOW 150 lb/sec
- GASPATH & COOLING-AIR TEMP. & PRESS.
- METAL TEMPERATURES AND HEAT FLUXES

AIR FLOW
P = 600 psia
T = 550°F-1160°F

COOLING AIR LINES
PLASMA SPRAYED ENVIRONMENT COATING

BUILT-IN GARDON TYPE HEAT FLUX SENSOR 0.060 in diam

LAMINATED TYPE HEAT FLUX GAGE 0.250 in diam AND AIRFOIL CURVATURE

LEAD WIRES

PLASMA SPRAYED CERAMIC MATERIAL

ELECTRON BEAM OR DIFFUSION BOND JOINING LINE

LASER OR ELECTRON BEAM WELD

IN-HOUSE METHODS FOR THE FABRICATION OF AIRFOILS WITH HEAT FLUX SENSORS
LOCAL HOT GAS TEMPERATURE AND HEAT FLUX INSTRUMENTATION
TYPICAL AIRFOIL AEROTHERMODYNAMIC ANALYSIS METHOD

HOT GAS BOUNDARY CONDITIONS

3-D BOUNDARY LAYER CODES
STAN5, FINITE DIFFERENCE
BLAYER, INTEGRAL

INVIScid
TSonic, 2D, BLADE TO BLADE
MERIDL, 2D, HUB TO SHROUD
MIT, 3D, (Thompkins)
Denton, 3D

VISCous
Nancy, 3D, (Dodge)
PEPSI, 3D, (McDonald)
MINT, 3D, (McDonald)
P. D. Thomas Beam-Warming Code, (UTSI)

BOUNDARY LAYER HEAT TRANSFER CODE

THREE-DIMENSIONAL STEADY STATE AND TRANSIENT CONDUCTION CODE

WALL TEMP. AND PRESS.

STRUCTURAL ANALYSIS CODE

LIFE PREDICTION CODE

COOLANT-SIDE HEAT TRANSFER CORRELATIONS

--INDUSTRY COOLANT-SIDE CORRELATIONS
--OPEN LITERATURE CORRELATIONS

ONE-, TWO-, OR THREE-DIMENSIONAL VISCOUS FLOW CODE OR NETWORK CODE

--INDUSTRY FLOW CODES
--TACT1, IMPINGEMENT INSERT
--FCFC, FULL COVERAGE FILM COOLED
--TACT2, MULTI-PASS
HOST IN-HOUSE HIGH TEMPERATURE TURBINE TESTING

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<tr>
<td>CONDUCT HEAT TRANSFER STUDIES OVER THE RANGE OF REAL-ENGINE CONDITIONS WITH NON-FILM COOLED AIRFOILS</td>
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<td>VERIFICATION TEST OF THE NASA BASELINE RESEARCH TURBINE WITH AIRFOIL METAL TEMPERATURE PREDICTION CODES</td>
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<td>MAP THE THREE-DIMENSIONAL FLOW FIELD IN A ROTATING PASSAGE WITH NO FILM COOLING</td>
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<td>MEASURE BI-AXIAL STRAIN DELTAS AT CRITICAL LOCATIONS ON STATIC AND ROTATING AIRFOILS AND THE DISK</td>
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<tr>
<td>VERIFICATION TEST OF ADVANCED TECHNOLOGY COOLED TURBINE WITH AIRFOIL METAL TEMPERATURE PREDICTION CODES</td>
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EXPECTED RESULTS

- REAL ENGINE VALUES OF LOCAL HOT GAS TEMPERATURES, LOCAL HOT GAS-SIDE HEAT TRANSFER COEFFICIENTS, AND LOCAL COOLANT-SIDE HEAT TRANSFER COEFFICIENTS FOR COMPARISON WITH ANALYTICAL MODELS AND CODES
- ASSESSMENT OF AIRFOIL METAL TEMPERATURE PREDICTION CODES WITH THE NASA BASELINE RESEARCH TURBINE
- VERIFY THE INVISCID FLOW REGION OF THE PASSAGE, WHICH WILL HAVE BEEN ESTABLISHED IN WARM TURBINE TESTS AND ESTABLISH THE ACTUAL VELOCITIES IN THE SECONDARY FLOW REGIONS OF THE PASSAGE FOR COMPARISON WITH FLOW CODES
- ASSESSMENT OF STRUCTURAL CODES WITH KNOWN THERMAL AND MECHANICAL BOUNDARY CONDITIONS
- ASSESSMENT OF AIRFOIL METAL TEMPERATURE PREDICTION CODES WITH AN ADVANCED TECHNOLOGY COOLED TURBINE