SPACE STATION EXPERIMENT DEFINITION:
ADVANCED POWER SYSTEM TEST BED

FINAL REPORT
15 December 1986

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NASA
National Aeronautics and Space Administration

Ford Aerospace & Communications Corporation
THE OBJECTIVE OF THIS EIGHT MONTH EFFORT WAS TO PROVIDE A CONCEPTUAL DESIGN FOR AN ADVANCED PHOTOVOLTAIC POWER SYSTEM TEST BED AND TO BETTER DEFINE THE REQUIREMENTS FOR ADVANCED PHOTOVOLTAIC POWER SYSTEM EXPERIMENTS. RESULTS OF THIS STUDY WILL BE USED IN THE DESIGN EFFORTS CONDUCTED IN PHASE B AND PHASE C/D OF THE SPACE STATION PROGRAM SO THAT THE TEST BED CAPABILITIES WILL BE RESPONSIVE TO USERS’ NEEDS. CRITICAL PV AND ENERGY STORAGE TECHNOLOGIES WERE IDENTIFIED AND INPUTS WERE RECEIVED FROM THE INDUSTRY (GOVERNMENT AND COMMERCIAL, U.S. AND INTERNATIONAL) WHICH IDENTIFIED EXPERIMENTAL REQUIREMENTS. THESE INPUTS WERE USED TO DEVELOP A NUMBER OF DIFFERENT CONCEPTUAL DESIGNS. PROS AND CONS OF EACH WERE DISCUSSED AND A STRAWMAN CANDIDATE IDENTIFIED. A PRELIMINARY EVOLUTIONARY PLAN, WHICH INCLUDED NECESSARY PRECURSOR ACTIVITIES, WAS ESTABLISHED AND COST ESTIMATES PRESENTED WHICH WOULD ALLOW FOR A SUCCESSFUL IMPLEMENTATION TO THE SPACE STATION IN THE 1994 TIME FRAME.
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1.0 Introduction

This report is the conclusion of a six month study, funded by NASA-Lewis Research Center, to provide a conceptual experimental design for the testing, development, and evaluation of advanced photovoltaic power system technologies on space station, and to document experimental requirements. Six primary tasks were identified which when completed would provide the required information. Ford has completed effort on all of these tasks (summarized in Section 2 below) and presented the inputs to NASA-Lewis at the three month and six month reviews. This report summarizes the findings of work performed on each of the tasks, provides a program cost summary and gives recommendations for future Test Bed efforts.
2.0 Summary of Program Tasks

Six major tasks were identified in the contract. The requirements of each of these tasks have been met and the findings are summarized below.

2.1 Task I - Identification of Critical Technologies

For this task, we (FACC) were to use our expertise in the area of power generation and storage to project state-of-the-art technologies at the end of the present decade and to identify critical photovoltaic power generation and power management and distribution technologies, and environmental interaction issues, for the growth space station. In addition, the following requirements were made:

- Provide rationale and justification for each technology identified and provide a proposed time-phased test plan for the Space Station.
- Identify specific experiments to be conducted, identify measurements required, and perform limited scaling analysis to establish suitable test bed size before incorporating the experiments into the test plan.

As part of the effort to identify 1990+ technologies, Ford sent out inquiry letters to over forty different companies and government agencies worldwide, asking them to identify technologies that they would like to see placed on the test bed and update the current status of those technologies. We received feedback from ten of these requests. These inputs, along with information we obtained as part of the Rocketdyne Space Station Work Package #4 team, were used to identify experiment requirements, scale the test bed and provide a time phased test plan for the Space Station. Table 1 summarizes the key experiments identified from these inputs. The results of these efforts were presented at the six week review. Summary details of this presentation, along with the inquiry mailing list are provided in Appendix 1. Also included are copies of the responses received.
Table 1
Inputs From Industry Regarding Use
of the Photovoltaic Test Bed

Aerospatiale
- Evaluation of annealing of GaAs cells
- Evaluation of Fresnel and Newton concentration systems (100x)
- Evaluate metal hydrogen battery operation in LEO
- Evaluate albedo affects on gridded back contact (GBC) cells
- Deployment and retraction testing of arrays which cannot be fully tested in 1 G.

United States Air Force
- Testing of Sodium Sulfur batteries

Jet Propulsion Laboratory
- On-orbit testing of lithium batteries

ARCO
- Qualification of tandem junction, amorphous cells

Mitsubishi
- On-orbit real time cell degradation evaluation
- Long term measurements of solar radiation (30 Years)
- On-orbit annealing characteristics of cells
- Evaluation of on-orbit repair techniques for large area solar arrays
- affects of plasma losses on high voltage arrays

Solarex
- Evaluation of long thermal cycling degradation of arrays
- Standard Cell Calibration

Sovonics
- Deployment/retraction testing of amorphous silicon arrays
- Radiation degradation/annealing of amorphous silicon arrays
- Plasma affects on amorphous silicon arrays
- evaluate limits of specific power capability of amorphous silicon arrays
- Evaluate the use of ion propulsion with amorphous silicon arrays
2.2 Task II - Determination of Experimental Requirements

The primary requirement of this task was to develop a conceptual design to identify resource requirements from the Space Station. Specific requirements that were addressed are shown summarized below:

- Provide sketches of the sub-scale test bed in sufficient detail to identify all modular and fixed elements, data acquisition devices and attachments to the Space Station or platform.
- Provide a functional diagram of changeouts and phases of the test program.
- Provide a list of all equipment necessary to perform various phases of the experiment.
- Provide physical characteristics including mass and specific dimensions.
- Identify requirements including orbit, viewing, power, thermal, data, crew, servicing and contamination sensitivities.
- Identify diagnostic and other instrumentation required to evaluate the environmental interaction.
- Provide a representative resource timeline which identify requirements for experiment start-up and assembly, operation, changeout, teardown, and stowage.

Each of the requirements above have been met. Detailed inputs for each of these requirements were presented at the three month review, held at NASA-Lewis on 3 February 1986. Table 2 is a summary of resource requirements for both major and minor experiments which have been identified to date. Table 3 identifies major components of the test bed. Additional summaries inputs relating to each of the above requirements are provided in Appendix 2.

2.3 Task III - Documentation of Experiment Requirements

The major requirement of this task was to provide the Mission Requirements Working Group (MRWG) and Technology Development Advocacy Group (TAG) with requirements of experiments which are planned for test bed integration. These requirements were to be inputted on the appropriate MRWG and TAG data forms. Initial inputs were to be provided at the three month review, with the final inputs provided at the close of the contract.
Table 2

RESOURCE REQUIREMENTS - SENSITIVITIES

<table>
<thead>
<tr>
<th>MAJOR EXPERIMENTS</th>
<th>VIEWING</th>
<th>POWER</th>
<th>THERMAL</th>
<th>DATA</th>
<th>CONTAMINATION</th>
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<tbody>
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<td></td>
<td>X</td>
<td>X</td>
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<tr>
<td>2. GROUNDING</td>
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<td>3. PV - CONCENTRATOR</td>
<td>SOLAR</td>
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<td>4. ENERGY WHEEL</td>
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<td>5. THERMOPHOTOVOLTAIC</td>
<td>SOLAR</td>
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<tr>
<td>6. BATTERY NaS</td>
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<td>X</td>
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<td>7. ANNEALING</td>
<td>SOLAR</td>
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<td>X</td>
<td>X</td>
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<td>8. BATTERIES - Li/TiS</td>
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<td>X</td>
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<td>9. BATTERY N1/H2</td>
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<td>10. MICROWAVE POWER</td>
<td>SOLAR</td>
<td></td>
<td></td>
<td>X</td>
<td></td>
</tr>
<tr>
<td>11. H2/Br REG FUEL CELL</td>
<td></td>
<td>X</td>
<td>X</td>
<td></td>
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</tr>
</tbody>
</table>

MINOR EXPERIMENTS

| 1. HIGH VOLTAGE PMAD |       | X     |         |      |               |
| 2. ALBEDO - GBC CELLS| SOLAR  |       |         | X    |               |
| 3. DEPLOY/RETRACT   |         |       |         |      |               |
| 4. PV - AMORPHOUS Si| SOLAR  |       | X       |      |               |
| 5. NEW TECH CELL QUAL| SOLAR |       | X       |      |               |
| 6. REAL TIME CELL DEG.| SOLAR |       | X       |      |               |
| 7. ON ORBIT REPAIR  |         |       |         |      |               |
| 8. STANDARDS CALIB. | SOLAR  |       | X       |      |               |

* EVA AND IVA SUPPORT REQUIRED TO INSTALL AND EXCHANGE EXPERIMENTS ON ABOUT THREE MONTH CENTERS AS SHOWN ON THE TIME-PHASED TEST PLAN.
This task has been completed. As part of the three month review, preliminary TAG and MRWG forms were presented for experiments which had been identified to date. The final copy of these forms, which identify requirements for all key experiments, will be delivered with this report.

Table 3

Major Components of the Test Bed

Mechanical
- Fixed pallet for ESS, PMAD, Thermal and Data
- Sun oriented platform or attachment
- RMS Capability
- Alpha and Beta joints for power/data transfer

Electrical
- Modularity of power converters for ESS management
- Solar array voltage regulation unit
- Power management and distribution units
- Slip rings across Beta joint
- Dissipative loads

Thermal
- Thermal fluid loop on fixed pallet
- Radiator surfaces
- Power supply and controller for resistance heaters
- Thermal blanket attachments
- Thermal instrumentation

Data/Control
- Microprocessor for Data, Test and Control
- DMS interface

Instrumentation
- Langmuir probes
- RF sounder for electron density
- Temperature
- Solar pointing angle
- Exposed and covered calibration cells

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2.4 Task IV - Detailed Conceptual Equipment Design

The conceptual design of the test bed was completed in this phase of the study. The requirements associated with this task are summarized below:

- Provide detailed sketches of equipment components to accomplish experimental objectives.
- Consider various attachment points for the test bed on the space station and platform options.
- Identify safety and interface issues.
- Identify automation and override features.
- Consider the advantages and disadvantages of having the test bed located on a free flyer, rather than the Space Station.

A detailed presentation of the efforts and results of this task was presented at the six month review. Figure 1 shows a baseline deployed test bed configuration. It could be located at a number of different location on the space station, each of which has its own attributes. These are summarized in Table 4. Figure 2 shows how this configuration would be stowed in the shuttle bay for launch. The automation and override features of the test bed were identified and a discussion of the pro's and con's of a free flyer test bed was made. Our recommendations are summarized in Table 5. Safety and interface issues were identified and solutions proposed. Table 6 summarizes the findings and provides recommended courses of action. Further details of each of the major efforts identified above are provided in Appendix 3.

Table 4

<table>
<thead>
<tr>
<th>Test Bed Site Selection Criteria</th>
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<tbody>
<tr>
<td>Solar Illumination Favored</td>
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<tr>
<td>Lower Boom</td>
</tr>
<tr>
<td>End of S/A Boom</td>
</tr>
<tr>
<td>Upper Boom</td>
</tr>
<tr>
<td>Upper Keel</td>
</tr>
<tr>
<td>Inner S/A Boom</td>
</tr>
<tr>
<td>Two-Part Design</td>
</tr>
</tbody>
</table>
END OF SOLAR ARRAY BOOM
CONFIGURATION STOWED IN SHUTTLE

NOTE: ALL DIMENSIONS IN METERS

APSTB STOWED CONFIGURATION

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Table 5

AUTOMATION AND OVERRIDE FEATURES

- RELEVANT EXPERIMENT REQUIREMENTS
  - ENERGY STORAGE EXPERIMENTS
    - μP CONTROLLED CHARGE/DISCHARGE CYCLING
    - OVERRIDE WITH BLOCK COMMANDS WHEREVER POSSIBLE
      - START UP
      - TURN OFF
      - INTERRUPT
  - SOLAR EXPERIMENTS
    - OCCASIONALLY MEASURE PERFORMANCE PARAMETERS
      (I.E. CURRENT-VOLTAGE CHARACTERISTICS)
  - PLASMA EXPERIMENTS
    - AUTOMATIC SEQUENTIAL COMMANDING

- OPERATIONS REQUIREMENTS
  - COMMANDS BY GROUND OR ASTRONAUT
  - TELEMETRY AVAILABLE TO ASTRONAUTS AND GROUND
  - COMPUTER MONITORIZATION OF CRITICAL PARAMETERS
    - ALARM WHEN LIMITS EXCEEDED
    - AUTOMATIC SAFING OF EXPERIMENTS
  - GROUND OR IVA OPERATIONS HANDBOOK AVAILABLE FOR CONTINGENCY OPERATIONS

2.5 Task V - Preliminary Evolutionary Plan

The major efforts associated with this task included the following:

- Identify major precursor activities and schedules which would be required of experiments in order to be ready for integration to the test bed.
- Identify shuttle flights that would be required to support these activities.
- Provide a schedule which identifies these activities and tasks by year, from the current state of each technology through integration on the test bed.
- Provide ROM costs by fiscal year for efforts that would result in the design, test and fabrication of the test bed, through integration on the space station.
Table 6

SAFETY - POTENTIAL HAZARDS

○ TEST BED HAZARDS:
  - HIGH TEMPERATURE RADIATORS
  - DUMMY LOAD RESISTIVE ELEMENTS
  - THERMAL CONTROL FLUID LOOPS
  - HIGH VOLTAGE HARDWARE

○ EXPERIMENT HAZARDS
  - ENERGY STORAGE EXPERIMENTS
    - HIGH TEMPERATURE RADIATORS (e.g. 350°C ON NaS BATTERY EXPERIMENTS)
    - POTENTIALLY EXPLOSIVE EXPERIMENTS (e.g. Li/TiS2 BATTERY AND ENERGY WHEEL)
  - SOLAR ARRAY EXPERIMENTS
    - HIGH VOLTAGE
    - REVERSE-BIASED SOLAR CELLS
    - NOMINALLY HIGH TEMPERATURE COMPONENTS
  - PMAD EXPERIMENTS
    - HIGH VOLTAGES

SAFETY - CATEGORIZATION OF HAZARDS

1. HAZARD RESULTS FROM TOUCH
   - HIGH TEMPERATURE SURFACES
   - HIGH VOLTAGE SURFACES

2. HAZARD OCCURS WHEN ASTRONAUT OBSTRUCTS RADIATOR VIEW OF DEEP SPACE

3. HAZARD OCCURS BY EXPERIMENT EXPLOSION

SAFETY PRECAUTIONS

CONDITIONS 1 OR 2.

- REMOTE LOCATIONS FOR HAZARDOUS HARDWARE
  - CONSCIOUS EFFORT REQUIRED TO ENTER POTENTIALLY HAZARDOUS AREA
  - EVA PRE-OPERATIONS CHECKLIST
  - WARNING LABELS

CONDITION 3.

- PROTECTIVE CONTAINERS TO DIRECT EXPLOSION IN SAFE DIRECTION
- LIMIT SIZE OR ELIMINATE IF SAFETY CANNOT BE ASSURED
- RECONFIGURE EXPERIMENT DURING EVA FOR SAVE OPERATION

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All of these tasks were completed. Inputs received from the commercial industry, and from NASA were used to identify major technologies which would most likely use the test bed for research and qualification. The status of each technology was then evaluated and maturity factor assigned. In addition, the experiments were rated as high, medium, or low priority for placement on the test bed. These inputs were then used to schedule necessary activity for each experiment between now and integration onto the test bed. This schedule identified precursor test requirements and shuttle flights necessary to bring the technologies up to a level which would allow them to use the test bed facility. For scheduling purposes, 1994 was used as the first available test bed date. Table 7 summarizes the findings for the high priority experiments. Schedules for the other experiments are included as part of Appendix 4.

In addition to the scheduling tasks, ROM costs by fiscal year were identified for the test bed design, fabrication and test efforts. A four phase design plan, similar to that being used for Space Station, was used in the costing. This plan is presented in Table 8. Overall costs to design, fabricate, test and integrate the test bed were estimated at approximately $32,000,000 (1986 dollars). Detailed summaries of component costs, as well as costs by fiscal year, are shown in Table 9 and Table 10 respectively. Additional inputs relating to this task can be found in Appendix 4.

2.6 Task VI - Scheduling and Reporting

The requirements associated with this task are summarized below:

- Perform a kick-off, six week, three month and six month review at NASA-Lewis.
- Provide monthly status reports to NASA-Lewis, including NASA 533P and 533M financial summary sheets
- Provide initial TAG and MRWG forms at the three month review.
- Provide final TAG and MRWG forms at the end of the contract.

All requirements for scheduling and reporting have been met. Three status update reviews were presented during the program at the NASA-Lewis facility. The final six month review was presented on 5 May 1986.
In addition to the in-person reviews, monthly reports were provided which summarized effort and costs to date. Cost summaries were provided on the NASA forms 533M and 533P as required.

Finally, initial TAG and MRWG forms were provided at the three month review. The final issue of these forms are presented along with this report which completes all reporting requirements.
Table 7.
PRECURSOR TEST PLAN / SCHEDULE
Time Frame (Year)  Test Bed
Available

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<td>1. Plasma</td>
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<td>2. High</td>
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</table>

A - Design and fab EM test item
B - Complete ground testing of EM test item
C - Design and fab Test flight unit
D - Shuttle flight for concept verification
E - Shuttle flight for testbed integration

PRIORITY PURPOSES
1- Top priority, must be flown
2- Important, will be flown
3- Low priority, space available
<table>
<thead>
<tr>
<th>TEST BED PHASES</th>
<th>OVERALL PHASES</th>
<th>OVERALL CONCEPTUAL TRADE STUDY</th>
<th>PHASE A</th>
<th>PHASE B</th>
<th>COMPONENT DESIGN TRADES</th>
<th>PRELIMINARY DESIGN</th>
<th>PHASE C</th>
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<th>PHASE D</th>
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<th>FAB</th>
<th>FAB FLIGHT MODEL (\text{Qual Upgrade?})</th>
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-15-
<table>
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<th>Component</th>
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<td>PRIMARY SOLAR ARRAY</td>
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</tbody>
</table>

**Subtotal** $14,950,000

<table>
<thead>
<tr>
<th>Component</th>
<th>Cost</th>
</tr>
</thead>
<tbody>
<tr>
<td>ASSEMBLY &amp; TEST</td>
<td>$1,500,000</td>
</tr>
<tr>
<td>HIGH RELIABILITY PARTS</td>
<td>$300,000</td>
</tr>
<tr>
<td>CONFIGURATION ENGINEERING</td>
<td>$1,000,000</td>
</tr>
<tr>
<td>SYSTEMS ENGINEERING</td>
<td>$700,000</td>
</tr>
<tr>
<td>MATERIALS ENGINEERING</td>
<td>$300,000</td>
</tr>
</tbody>
</table>

**Subtotal** $18,750,000

| G/A (17%)                        | $3,200,000 |

**Profit (12%)** $2,600,000

**Total Costs** $24,550,000
Table 10.

PHOTOVOLTAIC TEST BED ROM COST BY FISCAL YEAR (K$)

<table>
<thead>
<tr>
<th></th>
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</tr>
</thead>
<tbody>
<tr>
<td>PHASE B</td>
<td>150</td>
<td>75</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>PHASE C</td>
<td></td>
<td></td>
<td>500</td>
<td>5500</td>
<td>6000</td>
<td></td>
<td></td>
<td></td>
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<tr>
<td>PHASE D</td>
<td></td>
<td></td>
<td></td>
<td>7000</td>
<td>9000</td>
<td>3500</td>
<td></td>
<td></td>
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<tr>
<td>LAUNCH SUPPORT</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
<td>1000</td>
</tr>
<tr>
<td>TOTAL BY FISCAL YEAR</td>
<td>150</td>
<td>575</td>
<td>5500</td>
<td>6000</td>
<td>7000</td>
<td>9000</td>
<td>3500</td>
<td>1000</td>
</tr>
</tbody>
</table>

TOTAL ESTIMATED COST: $32,725,000
3.0 Other Efforts

Part of the output resulting from the in-person presentations were action items which FACC was asked to address. All of these were addressed and responses provided as part of the monthly reports. In addition, a summary of action item status was presented at the three month and six month reviews. The only action item discussed which was not an original requirement of the contract, was evaluating the possibility of including solar dynamics (SD) as part of the test bed facility. To address this issue, FACC contacted Work Package #4 team members at Rocketdyne to determine the current state of the technology. Sub-scale versions of the Brayton and Sterling SD systems are available in power levels of about one kilowatt. Initial indication is that it would be possible to include SD as part of the baseline test bed configuration. Additional studies on this option will have to be made in the next phase of this study to clearly define constraints and costs.

In addition to the contractual requirements of Section 2.0, Ford was asked to make two additional summary presentations to NASA agencies. On 14 July 1986, a summary of this report was reported to the Technology Development Advocacy Group (TAG), meeting at the Johnson Space Center. A similar presentation was made to NASA headquarters (Code R and Code S), Washington D.C., on 16 October 1986.
4.0 Conclusions and Recommendation

As a result of this study, Ford feels that having a dedicated Photovoltaic Test bed on the Space Station, or on a Free Flyer, is a desirable goal. Inputs from industry indicate their desire for such a test bed which would be available for technology verification and component qualification. Initial sizing and design of the test bed show that a number of different options and locations are possible which would provide the required capabilities. As part of Task V, Ford has provided a development plan, similar to that currently being used for Space Station, which would result in the integration of the Test Bed, onto the Space Station by 1994. The estimated costs for this capability are estimated to be about $32 million (1986 dollars). To help amortize these costs, Ford recommends that the test bed be considered as a possible revenue device where industry would pay for its use. This would not only reduce government costs, but provide for faster development and qualification of new technologies. Finally, Ford recommends that a follow-on, 'Phase B' effort be funded. Table 11 summarizes the tasks that should be completed during the Phase B effort.
Table 11.

Recommendations For Future Effort

- The overall test bed plan should follow a path similar to that being used for Space Station
- Phase 'B' efforts should be broken into two major tasks:
  (1) Component Design Trade Studies
     - Identify contractors for major test bed components
     - Perform cost/technical trades on each component
     - Select components to be used on the test bed
  (2) Preliminary Design
     - Complete design of testbed using components selected in (1) above.
     - Identify top priority experiments and interface with potential users
       - Verify experiment requirements
       - Agree upon responsibilities
       - Identify schedule/cost constraints
     - Customer (NASA) - Test Bed Interface
       - Verify location of test bed on space station
       - Interface with Space Station WP-2 contractor
     - Provide preliminary ICD's for the test bed to be used during Phase 'C' fabrication
APPENDIX 1.

- Request Letter
- Mailing List
- Copy of Responses
- Six Week Review Summary
Ford Aerospace &
Communications Corporation
Western Development
Laboratories Division

Phone: (415) 852-5137 or (415) 852-5131
3939 Fabian Way
Palo Alto, California 94303

October 31, 1985

Dear : 

Ford Aerospace and Communications Corporation (FACC) is currently involved in a study, funded by NASA Lewis, to help identify customer test needs for a photovoltaic power test bed which will be attached to the Space Station. The test bed will be designed using a solar array for primary power, an energy storage system and a power management and distribution (PMAD) system all of which will be isolated from the main Space Station power bus.

The objective of this study is to provide a conceptual design for this test bed as well as identify the type of experiments that might use such a facility. These experiments need not be specifically photovoltaic oriented, but may include other subsystems which interface with photovoltaics, such as energy storage systems and power conditioning equipment. By identifying requirements early enough, space station designers can be more responsive to potential users' needs. The targeted time frame for flight is 1992-1995 and the experiments to be conducted should be representative of 1990+ technologies.

As a result, FACC is asking for inputs, from solar cell manufacturers and other companies which interface with photovoltaic systems, that identify the type of experiments and technologies they perceive might use such a test bed, if it were available. The attached questionnaire may be used for inputting suggestions for such experiments and may be included as part of the final report to NASA identifying potential technologies, test bed requirements and test bed users. It would be greatly appreciated if this letter were forwarded to the appropriate personnel who may have inputs for this study.
The schedule for completion of this study is early 1986, so I ask that any ideas or inputs you may have be forwarded to the undersigned no later than 15 December 1985. Let me thank you in advance for your time and effort assisting with ideas for this study. Proper recognition of each contributor will be included as part of the study documentation. If you have any further questions, please feel free to contact me at the address or phone shown on the letterhead.

Sincerely,

FORD AEROSPACE & COMMUNICATION CORP.

Robert E. Neff  MS-G45
Power & Control Engineering Laboratory
MAILING LIST

Mr. Robert Annan
Director, Photovoltaics Activity
U.S. Department of Energy
1000 Independence Ave. SW
Washington D.C. 20585

Mr. Gerard Barkats
Solar Array Program Manager
Aerospatiale
100 Blvd du Midi
BP No 52-06322
Cannes-La-Bocca
France

Mr. John Barton
Electrical Power Systems
Boeing Aerospace Co.
P.O.Box 3999
Seattle, Wash. 98124

Mr. John Benner
Solar Energy Research Institute
1617 Cole Blvd.
Golden, Colorado 80401

A. O. Britting, Jr., M4-988
The Aerospace Corporation
P.O. Box 92957
Los Angeles, CA  90009

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Vice President and Marketing Manager
Applied Solar Energy Corporation
15251 E. Don Julian Road
P.O. Box 1212
City of Industry, Calif. 91749

Mr. Heiko Brodersen
Messerschmitt-Bolkow-Blohm
Unternehmensbereich - Raumfahrt
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8000 Munchen 80
West Germany

Rebecca Chaky
TRW
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One Space Park
Redondo Beach, CA 90278
Mickey Cornwall, MZ 24-6200
General Dynamics
Space Power Systems
P.O. Box 85357
San Diego, CA 92138

Bill Dunbar (3A-03)
Boeing Aerospace Company
P.O. Box 3999
Seattle, WA 98124

Gerald Fleck (R4/1128)
TRW Space & Technology Group
One Space Park
Redondo Beach, CA 90278

Dr. Robert Francis, M4-988
The Aerospace Corporation
P.O. Box 92957
Los Angeles, CA 90009

Ernie Frank, M4-988
The Aerospace Corporation
P.O. Box 92957
Los Angeles, CA 90009

Mr. Charles Gay
ARCO Solar, Inc.
P.O.Box 4400
Woodland Hills, Calif. 91365

Jack Geis
Aero Propulsion Laboratory
AFWAL/POOC-2
Wright-Patterson AFB, Ohio 45433-6563

Mr. Ron Given
Electrical Power Systems SSD
Lockheed Missiles & Space
1111 Lockheed Way
Sunnyvale, Calif. 94086

Mr. Leland Goldhammer
Hughes Aircraft Co.
7249 Berry Hill Drive
Rancho Palo Verdes, Ca. 90274
Mr. John Goldsmith  
Solarex  
201 Perry Parkway  
Gaithersburg, MD 20877

Captain David C. Hall  
HQ Foreign Technology Division/TQTD  
Wright-Patterson AFB, OH 45433-6508

Mr. Joseph Hanak  
Director, Photovoltaic Products  
Energy Conversion Devices, Inc.  
32400 Edwards Street  
Madison Heights, Mich. 48071

Joseph H. Hayden (512/V329)  
Power Electronics Department  
Hughes Aircraft Company  
P.O. Box 92919  
Los Angeles, CA 90009

Ed Horne (2R-00)  
Boeing Radiation Effects Laboratory  
1620 Henderson South  
Seattle, WA 98108

Mr. James Hutchby  
Research Triangle Institute  
P.O. Box 12194  
Research Triangle Park  
Durham, NC 27709

Dr. V. Josephson, M4/933  
The Aerospace Corporation  
P.O. Box 92957  
Los Angeles, CA 90009

Mr. Thomas Key  
Sandia National Laboratories  
P.O. Box 5800  
Albuquerque, NM 87185

Dr. Cedric Kitchen  
Manager, Space Technology  
Pilkington Electro-Optics Inc.  
Unit 2, Kinmel Park Industrial Estate  
Bodelwyddan, Rhyl, Clwyd, LL18 STY  
England

-26-
Mr. Hugh MacMillan  
Solar Cell Program Manager  
Varian  
3251 Olcott St.  
Santa Clara, Calif. 95054

Mr. Carl Mazzocco  
Manager, Power Generation Group  
General Electric  
Valley Forge Space Division  
King of Prussa, PA.

Mr. Robert Patterson  
TRW, Inc.  
One Space Park  
Redondo Beach, Calif  
90278

Mr. Phil Pierce  
RCA  
P.O. Box 800  
Princeton, NJ 08540

Dr. Kurt Roy  
Telefunken Electronics  
Solar Cell Department Manager  
Theresienstrasse 2  
D-7100  
Heilbronn, West Germany

Mr. Isadore Sachs  
Space Power & Thermal Control  
Optical Coating Laboratory, Inc.  
2789 Northpoint Parkway  
Santa Rosa, Calif. 95401-7397

Art Schoenfeld (R4/1104)  
TRW Space and Technology Group  
One Space Park  
Redondo Beach, CA 90278

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Jet Propulsion Laboratory  
4800 Oak Grove Drive M/S 125-231  
Pasadena, Calif. 91109

Neal F. Shepard, Jr.  
Spacecraft Operations, Bldg. 100, U2420  
General Electric Company  
P.O. Box 8555  
Philadelphia, PA 19101
Mr. Richard Statler  
Navel Reseach Laboratory  
4555 Overlook Ave. SW  
Code 6612  
Washington D.C. 20375

Mr. A.M.V. Vieleers  
Solar Array Program Manager  
Fokker B.V.  
P.O.Box 7600  
1117 ZJ Schiphol  
The Netherlands

Mr. Don Warnock  
Aerospace Propulsion Laboratory  
AFWAL/POOS-2  
Wright Patterson AFB, Ohio 45433-6563

Mr. Joseph Wise  
U.S. Air Force  
A.F. Wal/Poc  
Wright Patterson AFB  
Ohio 45433

Mitsubishi Electric Corporation  
c/o Mr. Carl Yamane  
P.O. Box 1650  
Palo Alto, CA 94302

Sharp  
c/o Mr. Carl Yamane  
P.O. Box 1650  
Palo Alto, CA 94302

Mr. Dieter Zemmrich  
Aerospace Programs Manager  
Spectrolab, Inc.  
12500 Gladstone Ave.  
Sylmar, Calif. 91342
INDUSTRY RESPONSES
December 16, 1985

Ford Aerospace & Communications Corp.
WDL Division
3939 Fabian Way
Palo Alto, CA 94303

Attention: Mr. Robert E. Neff, MS-G45
Power & Control Engineering Lab

Subject: Photovoltaic Test Bed for Space Station


Gentlemen:

We are pleased to enclose herewith Sharp's reply to the reference letter. Sharp has suggested five (5) types of experiments it believes useful for the photovoltaic test bed as seen on the attached questionnaires. We look forward to receiving your comments.

Very truly yours,

MITSUBISHI INTERNATIONAL CORP

Connie Lechnar
Contract Administrator
Aerospace Department
Palo Alto Office

/cc: MC TOK (MT-X)
enclosures
Mr. Robert E. Neff  
Ford Aerospace &  
Communications Corporation  
Western Development  
Laboratorics Division  
Power & Control Engineering Laboratory

Dear Mr. Neff:

Thank you for your letter, and it is our pleasure to propose for your request. Our ideas are presented for your Photovoltaic Power Test Bed which will be expected to fly in 1992-1995. We have 5 ideas for your request as stated on other sheets according to your format.

If you have any interest with our ideas and other solar power systems, or you start to design these experiments on your Photovoltaic Power Test Bed, we will propose to design ideas in detail for your request.

If you have any questions, please feel free to contact us at the address shown on the letterhead.

Sincerely,

SHARP CORPORATION

Akio Suzuki  
Division General Manager  
Photovoltaics Division  
Electronic Components Group
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: Evaluation of radiation degradation with various cells

PURPOSE: The objective is to compare and to evaluate the degradation of various solar cells for a long period in the same condition.

PROPOSED FLIGHT DATE (Year)

OPERATIONAL DAYS REQUIRED 30 years

APPROXIMATE MASS (Kg) 20

VOLUME:

STOWED: L x W x H = Cu.M.

DEPLOYED: L x W x H = Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): Solar

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0 Hrs/Day 0 No. of Days

OPERATIONS: 0 Hrs/Day 0 No. of Days

SERVICING: 0 Hrs/Day 0 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0 Hrs/Day 0 No. of Days

OPERATIONS: 0 Hrs/Day 0 No. of Days

SERVICING: 0 Hrs/Day 0 No. of Days

POWER REQUIREMENTS:

AVERAGE: _ KW PEAK: _ KW DC _ HZ

Hrs/Day: _ No. of Days:

DATA STORAGE: TAPE STORAGE IN-SITU TRANSMISSION NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO

OTHER REQUIREMENTS:

-32-
**EXPERIMENT TITLE:** High accuracy measurement of solar radiation

**PURPOSE:** The objective is to measure solar radiation and other radiations (for example, the distribution and intensity of earth's albedo) with very high accuracy for a very long term. (for example, 30 years)

**PROPOSED FLIGHT DATE (Year):**

**OPERATIONAL DAYS REQUIRED:** 30 years

**APPROXIMATE MASS (kg):** 20

**VOLUME:**

- **STOWED:** $L \times W \times H = \text{Cu.M.}$
- **DEPLOYED:** $L \times W \times H = \text{Cu.M.}$

**REQUIRED ORIENTATION (Inertial, solar, earth, other):** Solar & Earth

**EXTRA-VEHICULAR ACTIVITY REQUIRED:**

- **SET-UP:** 0 Hrs/Day 0 No. of Days
- **OPERATIONS:** 0 Hrs/Day 0 No. of Days
- **SERVICING:** 0 Hrs/Day 0 No. of Days

**INTRA-VEHICULAR ACTIVITY REQUIRED:**

- **SET-UP:** 0 Hrs/Day 0 No. of Days
- **OPERATIONS:** 0 Hrs/Day 0 No. of Days
- **SERVICING:** 1 Hrs/Day 60 No. of Days

**POWER REQUIREMENTS:**

- **AVERAGE:** _____ KW  PEAK: _____ KW
- **Hrs/Day:** _____ No. of Days: _____
- **AC:** _____ HZ

**DATA STORAGE:** TAPE STORAGE IN-SITU TRANSMISSION NONE

**DIRECT CONTROL FROM EARTH REQUIRED:** YES NO

**OTHER REQUIREMENTS:**

- 
- 
- 

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-33-
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: Regeneration test of solar cells by annealing

PURPOSE: The objective is to perform the regeneration test.

This experiment is to certify the technology which recovers degraded solar cells by annealing with laser or collected solar heat in space.

PROPOSED FLIGHT DATE (Year) ___________

OPERATIONAL DAYS REQUIRED ___10_____

APPROXIMATE MASS (Kg) __30__

VOLUME:

STOWED: L 1____ x W 1____ x H 0.3____ = 0.3 Cu.M.

DEPLOYED: L 1____ x W 1____ x H 1____ = 1 Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): Solar

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0 Hrs/Day 0 No. of Days

OPERATIONS: 0 Hrs/Day 0 No. of Days

SERVICING: 0 Hrs/Day 0 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 1 Hrs/Day 1 No. of Days

OPERATIONS: 1 Hrs/Day 2 No. of Days

SERVICING: 2 Hrs/Day 5 No. of Days

POWER REQUIREMENTS:

AVERAGE: ________ KW PEAK: ________ KW DC ________ Hz

Hrs/Day: ________ No. of Days: ________

DATA STORAGE: TAPE STORAGE_____ IN-SITU TRANSMISSION_____ NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES _____ NO ______

OTHER REQUIREMENTS: _________________________________
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: Engineering repair test of Large Solar Array

PURPOSE: The objective is to evaluate the repairing technique.
This technique is to find out and exchange the damaged part
of solar panels of a Large Solar Array in space.

PROPOSED FLIGHT DATE (Year) ________

OPERATIONAL DAYS REQUIRED 60

APPROXIMATE MASS (Kg) 100

VOLUME:

STOWED: L ______ x W ______ x H ______ = ______ Cu.M.

DEPLOYED: L ______ x W ______ x H ______ = ______ Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): Inertial

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 4 Hrs/Day 1 No. of Days

OPERATIONS: 4 Hrs/Day 40 No. of Days

SERVICING: 2 Hrs/Day 4 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 4 Hrs/Day 1 No. of Days

OPERATIONS: 4 Hrs/Day 40 No. of Days

SERVICING: 2 Hrs/Day 4 No. of Days

POWER REQUIREMENTS:

AVERAGE: _______ KW PEAK: _______ KW DC _______ Hz _______

Hrs/Day: _______ No. of Days: _______

DATA STORAGE: TAPE STORAGE _______ IN-SITU TRANSMISSION _______ NONE _______

DIRECT CONTROL FROM EARTH REQUIRED: YES _______ NO _______

OTHER REQUIREMENTS: ____________________________________________

______________________________________________________________
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: Investigation of interactions between a large solar array and space plasma

PURPOSE: The objective is to investigate the interactions between a large solar array with high bus-voltage and space plasma around the array for a long period.

PROPOSED FLIGHT DATE (Year) __________

OPERATIONAL DAYS REQUIRED 90

APPROXIMATE MASS (Kg) 100

VOLUME:

STOWED: L______ x W______ x H______ = _______ Cu.M.

DEPLOYED: L 30 x W 4 x H 0.5 = 60 Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): Solar

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0 Hrs/Day 0 No. of Days

OPERATIONS: 0 Hrs/Day 0 No. of Days

SERVICING: 0 Hrs/Day 0 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 1 Hrs/Day 1 No. of Days

OPERATIONS: 2 Hrs/Day 20 No. of Days

SERVICING: 0 Hrs/Day 0 No. of Days

POWER REQUIREMENTS:

AVERAGE: _______ KW PEAK: _______ KW DC _______ HZ

Hrs/Day: _______ No. of Days: _______

DATA STORAGE: TAPE STORAGE IN-SITU TRANSMISSION NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO

OTHER REQUIREMENTS: 

____________________________________

____________________________________

-36-
December 12, 1985

Mr. Robert E. Neff, MS-G45
Power and Control Engineering Laboratory
Ford Aerospace and Communications Corp.
3939 Fabian Way
Palo Alto, CA  94303

Dear Mr. Neff:

Enclosed please find responses to your October 31 letter.

Sincerely,

Edward M. Gaddy
Manager, Aerospace Products

EMG/bh

Enclosures
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: Solar Cell Calibration Experiment

PURPOSE: To calibrate solar cells to be used as primary standards in setting simulator outputs. A group of cells would be sent to space, protected from radiation by a thick aluminum cover. While pointed at the sun, the cover would be removed and the output of the cells measured. The cells are then recovered and returned to earth. This experiment would be very similar in purpose and structure to

(Cont. on the back)

PROPOSED FLIGHT DATE (Year) Once every other year

OPERATIONAL DAYS REQUIRED 1

APPROXIMATE MASS (Kg) 

VOLUME:

STOWED: L_______ x W_______ x H_______ = _______ Cu.M.

DEPLOYED: L_______ x W_______ x H_______ = _______ Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other):

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: ____ Hrs/Day ____ No. of Days

OPERATIONS: ____ Hrs/Day ____ No. of Days

SERVICING: ____ Hrs/Day ____ No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 1 Hrs/Day 1 No. of Days

OPERATIONS: 1 Hrs/Day 1 No. of Days

SERVICING: ____ Hrs/Day ____ No. of Days

POWER REQUIREMENTS:

AVERAGE: ____ KW PEAK: ____ KW DC ____ Hz

Hrs/Day: ____ No. of Days: ____

DATA STORAGE: TAPE STORAGE ____ IN-SITU TRANSMISSION ____ NONE ____

DIRECT CONTROL FROM EARTH REQUIRED: YES ____ NO ____

OTHER REQUIREMENTS:

_________________________________________________________________

_________________________________________________________________

-38-
EXPERIMENT TITLE: Solar Cell Thermal Cycling Experiment

PURPOSE: To test solar panel samples for their ability to withstand thermal cycling. Experiment would place solar panels on the space station for long term cycling. After being in space for years, the panels would be returned to earth for examination to determine effects of the exposure.

PROPOSED FLIGHT DATE (Year) ASAP

OPERATIONAL DAYS REQUIRED None

APPROXIMATE MASS (Kg) 3

VOLUME:

STOWED: \( L \times W \times H = 0.02 \) Cu.M.

DEPLOYED: \( L \times W \times H = \) Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): Ideally panel should face sun during sunlit period of the orbit.

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 1 Hrs/Day 5 No. of Days per year.

OPERATIONS: 1 Hrs/Day 5 No. of Days per year.

SERVICING: _____ Hrs/Day _____ No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: _____ Hrs/Day _____ No. of Days

OPERATIONS: _____ Hrs/Day _____ No. of Days

SERVICING: _____ Hrs/Day _____ No. of Days

POWER REQUIREMENTS:

AVERAGE: None KW PEAK: _____ KW DC _____ AC _____ HZ_____

Hrs/Day: _______ No. of Days: _______

DATA STORAGE: TAPE STORAGE None IN-SITU TRANSMISSION NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO

OTHER REQUIREMENTS:

-39-
REPLY TO
ATTN OF: AFWAL/POOS-2 (Don Warnock, (513) 255-6241) 17 December 1985

SUBJECT: Power System Experimentation Unit - Sodium-Sulfur Battery

TO: Ford Aerospace & Communications Corp.
    Western Development Laboratories Division
    Attn: Daniel L. Hutchins, MS-G45
    3939 Fabian Way
    Palo Alto, CA  94303

We would like to propose a sodium-sulfur battery experiment for consideration in your conceptual study of a power system experimentation unit as described in your letter of 26 November. Lt Ross Dueber of our group has filled out your questionnaire to the extent of our present knowledge. If you have any further questions please contact Lt Dueber at the same address or at telephone (513) 255-6241.

DON R. WARNOCK
Power Technology Branch
Aerospace Power Division
Aero Propulsion Laboratory

cc: AFWAL/POOS-2 (Lt Dueber)
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: Sodium-Sulfur Battery Flight Test

PURPOSE: To demonstrate the operability of a rechargeable Sodium-Sulfur battery in space environment.

PROPOSED FLIGHT DATE (Year) 1991 - 92

OPERATIONAL DAYS REQUIRED 140 days

APPROXIMATE MASS (Kg) 20 Kg

DIMENSIONS:
STOWED: L 43 cm x W 43 cm x H 32 cm
DEPLOYED: L 11' x W 11' x H 1

OTHER DIMENSIONAL CHARACTERISTICS:

REQUIRED ORIENTATION (Inertial, solar, earth, other):

POINTING ACCURACY REQUIRED: N/A

EXTRA-VEHICULAR ACTIVITY REQUIRED:
SET-UP: _______ Hrs/Day _______ No. of Days
OPERATIONS: _______ Hrs/Day _______ No. of Days
SERVICING: _______ Hrs/Day _______ No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:
SET-UP: _______ Hrs/Day _______ No. of Days
OPERATIONS: _______ Hrs/Day _______ No. of Days
SERVICING: _______ Hrs/Day _______ No. of Days

THERMAL REQUIREMENTS:
TYPE: OPEN CLOSED
OPERATING TEMPERATURE: 350 °C
HEAT DUTY
a) _______ watts
b) _______ watts

OTHER

POWER REQUIREMENTS:
AVERAGE: 100 KW PEAK: N/A KW DC
QUALITY: AC HZ
Hrs/Day: _______ No. of Days:

DATA STORAGE: TAPE STORAGE IN-SITU TRANSMISSION NONE
DIRECT CONTROL FROM EARTH REQUIRED: YES NO
RETRIEVEABILITY DESIRED: YES NO

OTHER REQUIREMENTS:

OTHER REQUIREMENTS:
M. Robert E. Neff  
FACC  
Western Development Laboratories Division  
3939 Fabian Way  
Palo Alto, California 94303

Le December 20th, 1985

Dear Bob,

In reply to your reference letter relative to a photovoltaic power test bed attached to the Space Station, I am enclosing herewith some suggestions which might be of some use in your prospective study.

For any further detail or clarification, please contact Lionel Pelenc, Tel 93.93.90.24 at our Cannes' establishment.

Sincerely yours

M. A. ZILIANI  
Solar Array Department Manager
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: Gallium Arsenide Cells Anealing

PURPOSE: Evaluate the possible annealing of Ga As cells with in orbit radiation fluence and under different temperature.

PROPOSED FLIGHT DATE (Year) 1942

OPERATIONAL DAYS REQUIRED 200

APPROXIMATE MASS (Kg) 3

VOLUME:
STOWED: L 0.5 x W 0.5 x H 0.5 = 0.125 Cu.M.

DEPLOYED: L 0 x W 0 x H 0 = 0 Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): Solar

EXTRA-VEHICULAR ACTIVITY REQUIRED:
SET-UP: 1 Hrs/Day 1 No. of Days
OPERATIONS: 0 Hrs/Day 0 No. of Days
SERVICING: 0 Hrs/Day 0 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:
SET-UP: 1 Hrs/Day 1 No. of Days
OPERATIONS: 1 Hrs/Day 200 No. of Days
SERVICING: 0 Hrs/Day 0 No. of Days

POWER REQUIREMENTS:
AVERAGE: 1 KW PEAK: 2 KW DC 30 V AC 30 Hz
Hrs/Day: 24 No. of Days: 200

DATA STORAGE: TAPE STORAGE IN-SITU TRANSMISSION NONE

DIRECT CONTROL FROM EARTH REQUIRED: yes no

OTHER REQUIREMENTS:

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PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: MICRO CONCENTRATION

PURPOSE: Evaluate degradation of micro concentration system in orbit, to correlate ground test
Micro concentrators will be cassegrarian, Fresnel lens
and Newton with 100 concentration factor.

PROPOSED FLIGHT DATE (Year) 1997 +

OPERATIONAL DAYS REQUIRED 30

APPROXIMATE MASS (Kg) 10

VOLUME:

STOWED: L 0.5 x W 0.5 x H 0.3 = 0.075 Cu. M.

DEPLOYED: L x W x H = Cu. M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): Solar

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 1 Hrs/Day 1 No. of Days
OPERATIONS: 0 Hrs/Day 0 No. of Days
SERVICING: 0 Hrs/Day 0 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 1 Hrs/Day 1 No. of Days
OPERATIONS: 1 Hrs/Day 30 No. of Days
SERVICING: 0 Hrs/Day 0 No. of Days

POWER REQUIREMENTS:

AVERAGE: 0.1 KW PEAK: 0.2 KW DC 30 V AC
Hrs/Day: 24 No. of Days: 30

DATA STORAGE: TAPE STORAGE X IN-SITU TRANSMISSION X NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO X

OTHER REQUIREMENTS:

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PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: METAL HYDROGEN BATTERY BEHAVIOUR

PURPOSE: The purpose of this experiment is to evaluate the behavior of a metal hydrogen battery on low orbit. This battery would consist typically in 2 pressurized nickel-hydrogen cells and an associated control unit.

PROPOSED FLIGHT DATE (Year) 1992

OPERATIONAL DAYS REQUIRED Throughout the flight (lifetime behaviour)

APPROXIMATE MASS (Kg) 4.5

VOLUME:

STOWED: L 0.2 x W 0.2 x H 0.2 = 8 x 10^-3 Cu.M.

DEPLOYED: L d° x W d° x H d° = d° Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): Baseplate toward North

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 1 Hrs/Day 1 No. of Days

OPERATIONS: / Hrs/Day / No. of Days

SERVICING: / Hrs/Day / No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: / Hrs/Day / No. of Days

OPERATIONS: / Hrs/Day / No. of Days

SERVICING: / Hrs/Day / No. of Days

POWER REQUIREMENTS:

AVERAGE: 0.035 KW PEAK: 0.50 KW DC 20A
AC 240V HZ

Hrs/Day: 16 No. of Days: All

DATA STORAGE: TAPE STORAGE X IN-SITU TRANSMISSION NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO X

OTHER REQUIREMENTS: TM/TC = 10

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EXPERIMENT TITLE: **INTEGRATED BIFACIAL CELLS**

**PURPOSE:** Measure power increase due to Earth albedo on rear side and its consequences at power control level.

PROPOSED FLIGHT DATE (Year) **1997**

OPERATIONAL DAYS REQUIRED **30**

APPROXIMATE MASS (Kg) **1**

**VOLUME:**

| STOWED: | L 0.3 x W 0.3 x H 0.7 = 0.018 Cu.M. |
| DEPLOYED: | L x W x H = Cu.M. |

REQUIRED ORIENTATION (Inertial, solar, earth, other): **Solar / earth**

**EXTRA-VEHICULAR ACTIVITY REQUIRED:**

| SET-UP: | 1 Hrs/Day 1 No. of Days |
| OPERATIONS: | 0 Hrs/Day 0 No. of Days |
| SERVICING: | 0 Hrs/Day 0 No. of Days |

**INTRA-VEHICULAR ACTIVITY REQUIRED:**

| SET-UP: | 1 Hrs/Day 1 No. of Days |
| OPERATIONS: | 1 Hrs/Day 30 No. of Days |
| SERVICING: | 0 Hrs/Day 0 No. of Days |

**POWER REQUIREMENTS:**

| AVERAGE: | 0.1 KW |
| PEAK: | 0.2 KW |
| DC: | 30 V |
| AC: | |
| HZ: | |
| Hrs/Day: | 24 |
| No. of Days: | 30 |

**DATA STORAGE:** TAPE STORAGE **X** IN-SITU TRANSMISSION **NONE**

**DIRECT CONTROL FROM EARTH REQUIRED:** **YES** NO **X**

**OTHER REQUIREMENTS:**

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PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: DEPLOYMENT / RETRACTION SYSTEM

PURPOSE: 1) 0g testing of deployable equipment not flexible on ground
          2) Life duration testing in vacuum and 0g conditions of motors

PROPOSED FLIGHT DATE (Year) 1997

OPERATIONAL DAYS REQUIRED 90

APPROXIMATE MASS (Kg) 15

VOLUME:
- STOWED: L 0.8 x W 0.5 x H 0.3 = 0.12 Cu.M.
- DEPLOYED: L 0.8 x W 2.0 x H 0.1 = 0.16 Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): Solar

EXTRA-VEHICULAR ACTIVITY REQUIRED:
- SET-UP: 1 Hrs/Day 1 No. of Days
- OPERATIONS: 0 Hrs/Day 0 No. of Days
- SERVICING: 1 Hrs/Day 10 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:
- SET-UP: 1 Hrs/Day 1 No. of Days
- OPERATIONS: 1 Hrs/Day 90 No. of Days
- SERVICING: 0 Hrs/Day 0 No. of Days

POWER REQUIREMENTS:
- AVERAGE: 0.1 KW
- PEAK: 0.2 KW
- DC 30 V
- AC _____ HZ
- Hrs/Day: 3
- No. of Days: 90

DATA STORAGE: TAPE STORAGE X IN-SITU TRANSMISSION X NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES X NO

OTHER REQUIREMENTS:

-
December 05, 1985

Robert E. Neff
Ford Aerospace & Communication Corp.
Western Development
Laboratories Division
3939 Fabian Way
Palo Alto, Calif. 94303

Dear Mr. Neff:

Attached is our preliminary response to your questionnaire regarding the proposed photovoltaic test bed for space station experimentation. We would like to be kept informed of progress on the project so that we can further refine our plans and expectations.

Yours very truly,

Don L. Morel
Director of Research
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: TANDEM THIN FILM PHOTOVOLTAIC MODULE

PURPOSE: SPACE QUALIFICATION

PROPOSED FLIGHT DATE (Year) 1992

OPERATIONAL DAYS REQUIRED 2

APPROXIMATE MASS (Kg) 4

VOLUME:

STOWED: \( L \times W \times H = 1.3 \times 0.61 \times 0.25 = 0.20 \) Cu.M.

DEPLOYED: \( L \times W \times H = 1.3 \times 0.61 \times 0.25 = 0.20 \) Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): SOLAR

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 1 Hrs/Day 1 No. of Days

OPERATIONS: 0 Hrs/Day 0 No. of Days

SERVICING: 0 Hrs/Day 0 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 1 Hrs/Day 1 No. of Days

OPERATIONS: 0 Hrs/Day 0 No. of Days

SERVICING: 0 Hrs/Day 0 No. of Days

POWER REQUIREMENTS:

AVERAGE: 0 KW PEAK: 0 KW DC AC HZ

Hrs/Day: 0 No. of Days: 0

DATA STORAGE: TAPE STORAGE ✓ IN-SITU TRANSMISSION NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO ✓

OTHER REQUIREMENTS: POWER RECORDING APPARATUS

-49-
December 4, 1985

Mr. Robert E. Neff, MS-G45
Power & Control Engineering Laboratory
Ford Aerospace & Communications Corporation
Western Development Laboratories Division
3939 Fabian Way
Palo Alto, California 94303

Dear Mr. Neff:

Thank you for your letter of October 31 in which you invited Energy Conversion Devices, Inc. (ECD) to propose tests for the Space Station. As you know ECD is engaged in the development of amorphous silicon alloy solar cells and modules through Sovonics Solar Systems, a partnership between ECD and the Standard Oil Company (SOHIO).

At Sovonics, during the past two years, we have developed the Ultragood® photovoltaic modules which may be suitable for applications in space. A detailed description of the structure and performance of these modules is included in two papers which I presented in the 18th IEEE Photovoltaic Specialists Conference in Las Vegas, October 21-25, of this year.

The tests that we propose involve such ultralight arrays. Together four tests are listed, involving three different arrays, each of which would have an output of about 1250 at AMO.

I should point out that we would be prepared to start such tests as early as 1987, so that our arrays would be in use rather than in tests on the Space Station. Please convey the message to NASA.

I look forward to collaboration with you in the near future.

Sincerely yours,

J. J. Hanak

JJH/sz

Enclosures (4)
PHOTOVOLTAIC POWER TEST BED DATA SHEET

DEPLOYMENT AND RETRACTION OF A ROLL-UP AMORPHOUS SILICON PV ARRAY*

I. EXPERIMENT TITLE: AMORPHOUS SILICON PV ARRAY

PURPOSE: (1) Feasibility of repeated deployment and retraction of a flexible roll-up PV array in and out of its cannister and its effect on array performance.

(2) Stiffness and vibration testing

(3) Temperature cycling

PROPOSED FLIGHT DATE (Year) 1992 or earlier

OPERATIONAL DAYS REQUIRED 30 to 90 days (See test II).

APPROXIMATE MASS (Kg) 50

VOLUME:

STOWED: L 2.7 x W 0.30 x H 0.15 = 0.12 Cu.M. PV Array Test Equipment

DEPLOYED: L 2.7 x W 0.5 x H 0.3 = 0.075 Cu.M. PV Array Test Equipment

REQUIRED ORIENTATION (Inertial, solar, earth, other): Solar

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0.5 Hrs/Day 1 No. of Days

OPERATIONS: 0 Hrs/Day 0 No. of Days - Operation can be automated; number of tests is optional.

SERVICING: 0 Hrs/Day 0 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0.5 Hrs/Day 1 No. of Days

OPERATIONS: 0.1 Hrs/Day 30-90 No. of Days - Monitoring automated experiment.

SERVICING: - Hrs/Day - No. of Days - As needed

POWER REQUIREMENTS:

AVERAGE: 0.5 KW DC

PEAK: 0.7 KW AC

Hrs/Day: 24 No. of Days: 30-90

HV 60

DATA STORAGE: TAPE STORAGE X IN-SITU TRANSMISSION X NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO X

OTHER REQUIREMENTS: NONE

* See references on reverse side.
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EFFECT OF LIGHT AND SPACE RADIATION ON THE

EXPERIMENT TITLE: PV PERFORMANCE OF ROLL-UP ARRAYS

PURPOSE: (The same experimental setup as in experiment I) amorphous silicon alloy PV cells after exposure to light, electron and proton radiation degrade somewhat in their PV performance, however, the performance can be restored upon 0.5-3 hour annealing at 150-200°C. In this experiment the array would be periodically retracted into its cannister and there annealed by solar heat to restore its performance.

PROPOSED FLIGHT DATE (Year) 1992 or earlier

OPERATIONAL DAYS REQUIRED 90

APPROXIMATE MASS (Kg) 50

VOLUME:

STOWED: L 2.7 x W 0.30 x H 0.15 = 0.122 Cu.M. PV ARRAY

DEPLOYED: L 2.7 x W 4.2 x H 0.05 = 0.57 Cu.M. PV ARRAY

TEST EQUIP.

0.5 0.7 0.3 = 0.122

0.5 0.5 0.3 = 0.57

REQUIRED ORIENTATION (Inertial, solar, earth, other):

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0.5 Hrs/Day 1 No. of Days

OPERATIONS: 0 Hrs/Day 0 No. of Days

SERVICING: 0 Hrs/Day 0 No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0.5 Hrs/Day 1 No. of Days

OPERATIONS: 0.1 Hrs/Day 10 No. of Days

SERVICING: - Hrs/Day - No. of Days AS NEEDED

POWER REQUIREMENTS:

AVERAGE: 0.5 KW PEAK: 0.7 KW DC

AC X

Hz 60

Hrs/Day: 24 No. of Days: 30-90

DATA STORAGE: TAPE STORAGE X IN-SITU TRANSMISSION X NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO

OTHER REQUIREMENTS: Helium or Argon gas, as a heat exchanger, would be desirable during the annealing experiments, approximately 200 liters total (at standard temperature and pressure).
III EXPERIMENT TITLE: HIGH VOLTAGE SOLAR ARRAY PLASMA PROTECTION TECHNIQUE

PURPOSE: Amorphous solicon alloy PV arrays are highly suitable for high voltage construction and operation. Because of their monolithic structure, the arrays are totally encapsulated in a flexible laminate. Because of this, protection against space plasma effects and high voltage operation are feasible. In particular, solution proposed by Dunbar ET AL can be easily incorporated. We propose deploying a continuous 1000 volt series-connected roll-up array 10 x 1 M in size, with taps at desired voltage intervals. Array voltage will be proportional to length deployed.

OPERATIONAL DAYS REQUIRED: 90

APPROXIMATE MASS (Kg): 40

STOWED: L = 1.2 x W = 0.40 x H = 0.20 = 0.10 Cu.M. PV ARRAY

DEPLOYED: L = 10.3 x W = 1.1 x H = 0.05 = 0.57 Cu.M.

REOUIRED ORIENTATION (Inertial, solar, earth, other): SOLAR

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0.5 Hrs/Day 1 No. of Days
OPERATIONS: 0 Hrs/Day 0 No. of Days
SERVICING: _____ Hrs/Day No. of Days Not anticipated.

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0.5 Hrs/Day 1 No. of Days
OPERATIONS: 2 Hrs/Day 10 No. of Days
SERVICING: _____ Hrs/Day No. of Days As Needed

POWER REQUIREMENTS:

AVERAGE: 0.5 KW PEAK: 0.7 KW DC
Hrs/Day: 24 No. of Days: 90 (life testing at progressively higher voltage.)

DATA STORAGE: TAPE STORAGE X IN-SITU TRANSMISSION X NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO

OTHER REQUIREMENTS: NONE

* See references on reverse side.

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From: J. J. Hanak
Sovonics Solar Systems

PHOTOVOLTAIC POWER TEST BED DATA SHEET

LIMIT OF SPECIFIC POWER OF ULTRALIGHT

I. EXPERIMENT TITLE: AMORPHOUS SILICON PV ARRAYS

PURPOSE: Amorphous silicon alloy PV modules have already reached specific power of 2,400
\( \text{W/kg} \) at \( \text{AM1} \) (Ref. 1) and 5,000 \( \text{W/kg} \) is expected. We propose to construct a series of arrays
having progressively lower mass and module thickness, to determine the maximum practical array
specific power. The results of these tests have important implications for interplanetary
spacecraft, ion-propelled spacecraft and for fuel consumption of LEO spacecraft.

PROPOSED FLIGHT DATE (Year) 1992 or earlier

OPERATIONAL DAYS REQUIRED 90

APPROXIMATE MASS (Kg) 50

VOLUME:

STOWED: \( \begin{array}{ccc} L & 2.7 & \text{m} \\ W & 4.2 & \text{m} \\ H & 0.05 & \text{m} \end{array} \) = 0.075 Cu.M. Array

DEPLOYED: \( \begin{array}{ccc} L & 2.7 & \text{m} \\ W & 0.3 & \text{m} \\ H & 0.15 & \text{m} \end{array} \) = 0.122 Cu.M. Array

REQUIRED ORIENTATION (Inertial, solar, earth, other): Solar

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 2.0 Hrs/Day 1 No. of Days

OPERATIONS: 0 Hrs/Day 0 No. of Days

SERVICING: ___ Hrs/Day ___ No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: 0.5 Hrs/Day 1 No. of Days

OPERATIONS: 0.1 Hrs/Day 10 No. of Days - Over 90 days

SERVICING: ___ Hrs/Day ___ No. of Days

POWER REQUIREMENTS:

AVERAGE: 0.5 kW PEAK: 0.7 kW DC

Hrs/Day: 24 No. of Days: 90

AC ___ X ___ HZ 60

DATA STORAGE: TAPE STORAGE IN-SITU TRANSMISSION NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO

OTHER REQUIREMENTS: ________________________________

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PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: ION PROPULSION POWERED BY AMORPHOUS SILICON PV ARRAY

PURPOSE: BECAUSE OF VERY HIGH SPECIFIC POWER, a-Si ULTRALIGHT ARRAYS WOULD BE IDEAL FOR INTERPLANETARY SPACECRAFT PROPELLED BY ION THRUSTERS. A STATIC TEST OF ARBITRARY SIZE IS PROPOSED FIRST ON THE SPACE STATION TO TEST THIS CONCEPT. LATER A TEST OF SPACECRAFT LAUNCHED FROM THE SPACE STATION IS PROPOSED.

PROPOSED FLIGHT DATE (Year) 1992-1995

OPERATIONAL DAYS REQUIRED

APPROXIMATE MASS (Kg)

VOLUME:

STOWED: L_______ x W_______ x H_______ = _______ Cu.M.

DEPLOYED: L_______ x W_______ x H_______ = _______ Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): ________________

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: _____ Hrs/Day _____ No. of Days

OPERATIONS: _____ Hrs/Day _____ No. of Days

SERVICING: _____ Hrs/Day _____ No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: _____ Hrs/Day _____ No. of Days

OPERATIONS: _____ Hrs/Day _____ No. of Days

SERVICING: _____ Hrs/Day _____ No. of Days

POWER REQUIREMENTS:

AVERAGE: _______ KW PEAK: _______ KW DC _____ AC _____ HZ_____ 

Hrs/Day: _______ No. of Days: _______

DATA STORAGE: TAPE STORAGE _______ IN-SITU TRANSMISSION _______ NONE_____

DIRECT CONTROL FROM EARTH REQUIRED: YES____ NO____

OTHER REQUIREMENTS: ____________________________________________

-55-
Mr. Robert E. Neff MS-G45  
Power and Control Engineering Laboratory  
Ford Aerospace and Communication Corp.  
3939 Fabian Way  
Palo Alto, CA 94303 


Dear Mr. Neff:  

In response to your referenced letter, JPL Section 342, the Electric Power Systems Section, would be pleased to propose an experiment for a space station photovoltaic power test bed if it existed. The title of the experiment we would like to undertake is "Rechargeable Lithium Battery Power System." 

As a background, JPL, as the NASA Lead Center for lithium high energy cell technology, has been working over the past five years to develop lithium-base cell/batteries, both primary and secondary. The program has advanced to the point where test specimens have shown promise of becoming safe, reliable, high energy, primary and secondary power sources. The specific program objectives for the primary and secondary power sources are as follows: 

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<tr>
<th>OBJECTIVE</th>
<th>GOAL</th>
<th>APPLICATIONS</th>
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<tr>
<td><strong>Li-SOCl₂ PRIMARY</strong></td>
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<tr>
<td>Increase Specific Energy</td>
<td>300 WH/Kg</td>
<td>o Probes, Penetrators</td>
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<tr>
<td>High Rate Capability</td>
<td>2 H</td>
<td>o Vehicles</td>
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<td>Increase Storage Life</td>
<td>5 Years</td>
<td>o Free Flyers/Short Term</td>
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<td>Cylindrical/Prismatic Cells</td>
<td>10-500 AH</td>
<td>o Maneuvering Equip</td>
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<td>Space</td>
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<td>o Crew Equipment</td>
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<td>Station</td>
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<td><strong>Li-TiS₂ SECONDARY</strong></td>
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<tr>
<td>Increase Specific Energy</td>
<td>100 WH/Kg</td>
<td>o Planetary</td>
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<tr>
<td>Demonstrate Cycle Life</td>
<td>100000 Cycles</td>
<td>o Comet Rendezvous</td>
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<tr>
<td>Extend Life</td>
<td>10 Years</td>
<td>o Rovers</td>
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<td></td>
<td></td>
<td>o Planetary Stations</td>
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Since the lithium-base battery technology is presently in the research/development stage and is expected to mature with experimental flight-worthy cells/batteries early in the 1990's, the proposed experiment should meet the requirements for the test bed program planning.

The proposed experiment will require the photovoltaic source in order to evaluate the performance of the secondary battery as a viable rechargeable power source under the actual zero g gravity environment. The power from the photovoltaic source will be used to charge the battery during sunlight periods and the battery will be discharged to load during the sun occultation periods. The experiment will contain the secondary battery, the associated electronics for discharging and charging the battery, the load, and telemetry measurements of battery voltage, current and temperature. The objective of the experiment is to evaluate the performance and life electrochemical parameters of this high energy secondary lithium base battery and the required electronics to support its performance as a power source (charging and discharging cycles) at the actual space operating conditions (PV source, 0 g gravity). The block diagram of the experiment is shown below.

JPL will be planning to fabricate, test and deliver a flight worthy experiment to meet the requirements and schedules of the PV power test bed program. The data supplied to the attached questionnaire should be considered preliminary. If you have any further questions, please feel free to contact me at the address or phone shown.

Sincerely,

E. N. Costogoue, Member of Technical Staff
Space Power Section
(818) 354-3922
Mail Stop 198-220

cc: W. Bachman
    R. Key
    H. Stadler
    R. Stephenson
PHOTOVOLTAIC POWER TEST BED DATA SHEET

EXPERIMENT TITLE: Rechargeable Lithium Battery Power System

PURPOSE: Evaluate the performance of a high specific energy rechargeable power source (lithium base battery) in LEO flight environment with associated power hardware to assess the feasibility of becoming a viable secondary power source.

PROPOSED FLIGHT DATE (Year) 1992 or later

OPERATIONAL DAYS REQUIRED 1-2 years

APPROXIMATE MASS (Kg) 10.5 kg

VOLUME:

STOWED: L x W x H = .007 Cu.M.

DEPLOYED: L x W x H = .007 Cu.M.

REQUIRED ORIENTATION (Inertial, solar, earth, other): N/A

EXTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: N/A Hrs/Day N/A No. of Days

OPERATIONS: N/A Hrs/Day N/A No. of Days

SERVICING: N/A Hrs/Day --- No. of Days

INTRA-VEHICULAR ACTIVITY REQUIRED:

SET-UP: N/A Hrs/Day N/A No. of Days

OPERATIONS: 24 Hrs/Day 365+ No. of Days

SERVICING: N/A Hrs/Day --- No. of Days

POWER REQUIREMENTS:

AVERAGE: 150 watts KW

PEAK: N/A KW

Hrs/Day: 24 No. of Days: 365+

OR

DATA STORAGE: TAPE STORAGE X IN-SITU TRANSMISSION X NONE

DIRECT CONTROL FROM EARTH REQUIRED: YES NO X

OTHER REQUIREMENTS:

1-Remove the experiment if failure of more than two battery cells is noted.

2-Temperature control, battery operating range, 10-30°C.
SUBJ.: CONCEPTIONAL DESIGN AND POSSIBLE TEST NEEDS FOR A PHOTOVOLTAIC POWER TEST BED

DEAR MR. NEFF,

IN ADVANCE THE INFORMATION FOR YOU FOR EXPERIMENTS IN SPACE. WE ARE INTERESTED IN THE FOLLOWING THREE EXPERIMENTS CALLED

- BIFACIAL CELL FOR THE SPACE
  IT SHALL BE PROVEN THE ADVANTAGE OF THIS CELL COMPARED TO THE CONVENTIONAL CELL. IT USES THE ALBEDO-RADIATION. THEREFORE THE EFFECTIVE EFFICIENCY OF THIS CELL TYPE IS HIGHER. HOWEVER IT CAN BE MORE SENSITIVE AGAINST ELECTRON AND PROTON IRRADIATION. WHERE THE BIFACIAL CELL'S POWER OUTPUT AND THE HIGHER DEGRADATION RATE SHALL BE DETERMINED.

- INFLUENCE OF PROTON IRRADIATION ON THE PN-JUNCTION OF SILICON SOLAR CELLS
  THE INFLUENCE OF THE PROTON IRRADIATION ON THE ELECTRICAL PARAMETERS OF SOLAR CELLS SHALL BE INVESTIGATED FOR DIFFERENT PN-STRUCTURES.

- NEW CELL STRUCTURES FOR THE SPACE
  THE ELECTRICAL CHARACTERISTICS AND THE DEGRADATION BEHAVIOUR OF THE GaAs SOLAR CELL AND/OR OF THE INVERSION LAYER-SOLAR CELL AND/OR THE AMORPHOUS SOLAR CELL SHALL BE INVESTIGATED.

THE FOLLOWING MEASUREMENTS AND ANALYSIS ARE NECESSARY

- CURRENT-VOLTAGE-CHARACTERISTICS
- CELL TEMPERATURE
- INCIDENT SOLAR ENERGY
- INCIDENT ELECTRON AND PROTON IRRADIATION (DOSIS AND DENSITY)
- FOR THE P-GRID CELL IT IS DESIRABLE IF THE CELL COULD BE ROTATED.

FOR THE EXPERIMENTS WE NEED A SPACE OF 60 CM X 100 CM. THE MASS OF THE SOLAR GENERATORS WILL ABOUT 500 - 600 KGS.

IN JANUARY YOU WILL GET THE FILLED AP FORMULA AND SOME MORE DETAILS.

I HOPE THESE INFORMATIONS ARE USEFUL FOR YOUR FIRST STUDY.

WE BEST REGARDS AND BEST WISHES FOR THE NEW YEAR.

TELEFUNKEN ELECTRONIC
7100 HEILBRONN
DR. ROY-SR

IF ANY FURTHER DISTRIBUTION IS NECESSARY PLEASE NOTIFY CMC AT EXT 7662

-59-
Mr. Robert E. Neff,
Engineering Specialist - Space Systems,
FORD AEROSPACE & COMMUNICATIONS CORP.,
Western Development, Laboratories Division,
3939 Fabian Way,
PALO ALTO,
California 94303
United States.

26th February, 1986.

Dear Mr. Neff,

In response to your letter of 31st October 1985 to Dr. C. Kitchen regarding our likely requirements for a photovoltaic power test bed as part of the Space Station overall concept, I have the pleasure of putting forward some of Pilkington Space Technology's ideas and concepts.

The concepts that we are suggesting are, we feel, representative of 1990's technology and therefore are hard to quantify onto the summary sheets at this stage. In general terms the experiments we would propose submitting in the period 1992-95 are:-

(1) Concentrators - this would involve large area, ultra light reflectors either passive or active e.g., solar dynamic. Alternatively, this may involve diffractive optics - e.g., large area holographic lenses, as a means of concentration. A panel area of several (approx 5) square meters would be used with a mass of a number of kilogrammes. The experiment would be passive, i.e., no activity apart from deployment would be required. The performance of the modules would be monitored and recorded.

(2) Solar Cell Protection - this will involve alternative methods of bonding coverglasses to solar cells. Possible methods include FEP Teflon and electrostatic bonding and also RF sputtering. In addition, alternative materials to glass may also be available for testing.

The experiment would utilise a small panel area of approximately 1m² and only require deployment and periodic inspection. The performance would be monitored and the cell output recorded.

Continued......
(3) Solar Cell Annealing - the possibility of using laser annealing to regenerate the solar cells following damage by radiation bombardment. This may require alternative types of coverglass. As in (2) the experiment would require a small panel area of approximately 1m². Following a period in orbit after deployment, the panel would be subjected to laser annealing - the performance being monitored throughout.

At present these three concepts are the most obvious to us. However, I am confident that in the intermediate time between now and the 1990's other technological experiments and opportunities will become apparent.

We would welcome any further comments or information that you may have and would wish to be actively involved in any subsequent discussions or work studies.

Yours sincerely,

Peter White
Project Engineer.
SIX WEEK REVIEW SUMMARY

SPACE STATION EXPERIMENT DEFINITION:

ADVANCED POWER SYSTEM TEST BED

6 DECEMBER 1986
<table>
<thead>
<tr>
<th>INTERESTED ORGANIZATIONS</th>
<th>PURPOSE</th>
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<tbody>
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<td>I</td>
<td>CD</td>
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<td>N D N J</td>
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<td>T A</td>
<td>G L E G D</td>
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<td>R</td>
<td>H P H E</td>
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**POWER GENERATION**

<table>
<thead>
<tr>
<th>Planar Array</th>
</tr>
</thead>
<tbody>
<tr>
<td>1) High voltage effects (Plasma Interactions)</td>
</tr>
<tr>
<td>2) Albedo utilization</td>
</tr>
<tr>
<td>3) Flex blanket</td>
</tr>
</tbody>
</table>

**Concentrator Array**

| 1) TRW Cassegranian |
| 2) GE |
| 3) GD Slats |

**Solar Cells**

| 1) GaAs |
| 2) Multiple Junction |
| 3) Plasmon |
| 4) Albedo Utilization |

**ENERGY STORAGE**

| 1) Li/TiS2 Battery |
| 2) NiH2 Bipolar |
| 3) NaS Battery |
| 4) H2/X2 RFC |
| 5) N2H4 APU |
| 6) Flywheels |

**POWER MANAGEMENT & DISTRIBUTION**

| 1) Testing of High Voltage/High Power Handling Equipment |
| 2) Electromagnetic Interference |
| - Static Charge arc Discharge |
| 3) Microwave Power Transfer |

- X X

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EXPERIMENT 1. SOLAR ARRAY, CONVENTIONAL

OBJECTIVE - HIGH VOLTAGE ARRAY - PLASMA INTERACTION

TEST ITEM - PLANER SOLAR ARRAY, SS TECHNOLOGY

APPROXIMATE SIZE - PANEL SIZE 0.366 X 4.2 M
- PANEL VOLTAGE 80 VDC

14 PANELS = 1000 VOLTS, THEREFORE, MINIMUM SIZE
4.2 X 5.1 M AND PRODUCES 2.3 KW

TWO ARRAYS PROVIDE CONTROL AND VARIABLE SET-UP

TEST VARIABLES - ARRAY VOLTAGE, +/- 1000 VDC
- SUBSTRATE GROUNDING

MEASUREMENTS - PLASMA DENSITY, ARRAY I AND V

EXPERIMENT 2. POWER SYSTEM GROUND

OBJECTIVE - MINIMIZE PLASMA CHARGE ON PV POWER SYSTEM
- INCREASE POWER SYSTEM EFFICIENCY AND EVA SAFETY

TEST ITEM - TEST BED PV ARRAY - ESS - PMAD

APPROXIMATE SIZE - TEST BED DEFINED IN EXPERIMENT 1
- ABLE TO OBTAIN +/- 1000 VDC

TEST VARIABLES - ISOLATION OR CONNECTED GROUND
- POSITIVE, NEGATIVE, INTERMEDIATE

MEASUREMENTS - PLASMA DENSITY
- POWER SYSTEM OPERATING CHARACTERISTICS
EXPERIMENT 3. CONCENTRATOR ARRAY

OBJECTIVE - CONCENTRATOR TECHNOLOGY VERIFICATION

TEST ITEM - TRW MINIATURE CASEGRANIAN
- GE AND GD ALTERNATE DESIGNS

APPROXIMATE SIZE - 40 SUBPANELS OF 0.7 X 0.7 M PANEL

CONFIGURATION 3 X 7.6 M AND PRODUCES 2.3 KW

ALTERNATE ARRAY CONTENTIONAL PLANER FOR COMPARISON

TEST VARIABLES - VOLTAGE VS PLASMA
- POINTING
- STABILITY OF OPTICCS
- STABILITY, PLANARITY OF STRUCTURE
- ADVANCED SOLAR CELL CONFIGURATION

MEASUREMENTS - SUN SENSORS TO RESOLVE POINTING OF STRUCTURE
- ELECTRICAL OUTPUT, PLASMA DENSITY

EXPERIMENT 4. ENERGY WHEEL

OBJECTIVE - PERFORMANCE VERIFICATION

TEST ITEM - COUNTGER ROTATING AFLYWHEEL ASSEMBLY, 4 KW UNIT

APPROXIMATE SIZE - A 4 KW UNIT IS 0.5 X 0.5 X 1.0 METER
- MASS ABOUT 200 KG WITHOUT ELECTRONICS

TEST VARIABLES - NORMAL CHARGE/DISCHARGE
- PEAK LOAD CAPABILITY
- LIFETIME IN ZERO GRAVITY
- ADVERSE OPERATION

MEASUREMENTS - RESULTANT FORCES TO TEST BED
- INPUT/OUTPUT ELECTRICAL CHARACTERISTICS

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EXPERIMENT 5. AMORPHOUS SILICON ARRAY

OBJECTIVE - ENVIRONMENTAL STABILITY

TEST ITEM - INTEGRAL AROLL UP CELL ASSEMBLY

APPROXIMATE SIZE - 2 M WIDE, 10 M LONG BLANKET TO DELIVER 1.4 KW, ESTIMATED WEIGHT WITH DEPLOYMENT UNIT IS 400 KG

TEST VARIABLES - DEPLOYMENT/RETRACTION
- HIGH TEMPERATURE ANNEALING

MEASUREMENTS - ARRAY CURRENT, VOLTAGE, TEMPERATURE
- PLASMA DENSITY

EXPERIMENT 6. HIGH ENERGY DENSITY/CHARGE EFFICIENCY NaS BATTERY

OBJECTIVE - NaS BATTERY PERFORMANCE VERIFICATION
- WICKING OF MOLTEN Na TO ELECTROLITE
- WICKING OF MOLTEN S/Na POLYSULFIDE

TEST ITEM - NaS CELLS AND THERMAL CONTROL
- INTERFACE WITH TEST BED POWER SYSTEM

APPROXIMATE SIZE - 0.7 M DIA X 0.5 M HIGH WITH ESTIMATED MASS OF 100 KG, FOR 2 KW

TEST VARIABLES - COLD LAUNCH, 350 C OPERATION
- MULTIPLE CHARGE-DISCHARGE CYCLES

MEASUREMENTS - CHARGES/DISCHARGE CURRENT AND VOLTAGE
- TEMPERATURE
EXPERIMENT 7. HIGH SPECIFIC ENERGY, Li/TiS₂ BATTERY

OBJECTIVE - PERFORMANCE VERIFICATION
  - ZERO GRAVITY ELECTROLYTE BEHAVIOR

TEST ITEM - SELF CONTAINED ACELL MODULE, JPL

APPROXIMATE SIZE - 1.0 x 0.5 x 0.5 m with mass 100 kg, 1 kW unit plus large radiator and thermal system

TEST VARIABLES - MULTIPLE CHARGE/DISCHARGE CYCLES

MEASUREMENTS - CURRENT, VOLTAGE, TEMPERATURE

EXPERIMENT 8. HYDROGEN-BROMINE REGENERATIVE FUEL CELL

OBJECTIVE - BROMINE PERFORMANCE IN ZERO GRAVITY
  - ADEQUATE CONTACT OF HBr/Br₂ TWO PHASE MIXTURE WITH METALIC AND SEPARATOR SURFACES
  - SEPARATION AND STORAGE OF Br₂ AND HBr
  - DATA BASE FOR HIGH EFFICIENCY, 80% SYSTEM

TEST ITEM - FUEL CELL WITH ACTIVE THERMAL CONTROL

APPROXIMATE SIZE - 1.0 x 0.5 x 0.5 m, mass 200 kg for 2 kW unit

TEST VARIABLES - MULTIPLE CHARGE/DISCHARGE CYCLES

MEASUREMENTS - CURRENT, VOLTAGE, PRESSURE, ELECTROLYTE CONCENTRATION

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EXPERIMENT 9. NiH2 Bipolar Battery

OBJECTIVE - PERFORMANCE VERIFICATION OF COMPACT, MODULAR UNIT
- REDISTRIBUTION OF ELECTROLYTE AT LAUNCH

TEST ITEM - NiH2 Bipolar Battery with Active Cooling
APPROXIMATE SIZE - 1.5 X 0.4 X 0.4 M, MASS 300 KG

5 KW UNIT, PLUS THERMAL CONTROL

TEST VARIABLES - MULTIPLE CHARGE/RECHARGE CYCLES
MEASUREMENTS - CURRENT, VOLTAGE, TEMPERATURE, PRESSURE,
COOLING FLOW RATE

EXPERIMENT 10. High Voltage/Power PMAD

OBJECTIVE - DEMONSTRATION OF PERFORMANCE IN PLASMA
- ATTRACTION OF SPACE DEBRIS WITH RESULTING
SHORT OF HIGH VOLTAGE CIRCUITS

TEST ITEM - High Voltage/High Power Charge Converter
APPROXIMATE SIZE - 0.3 X 0.5 X 0.6 M, MASS 30 KG FOR 5 KW UNIT

TEST VARIABLES - OPERATING VOLTAGE AND EXPOSURE TIME
MEASUREMENTS - ELECTRICAL PARAMETERS UNDER LOAD
EXPERIMENT 11. MICROWAVE POWER TRANSFER

OBJECTIVE - TEST EFFICIENCY OF POWER TRANSFER FROM STATION TO PLATFORM
- DETERMINE RF FIELD INTERACTIONS

TEST ITEM - HIGH POWER-HIGH FREQUENCY TRANSMITTER, ANTENNA AND RECEIVER

APPROXIMATE SIZE - 1 TRANSMITTER, 0.3 X 0.3 X 0.6 M, 50 KG
2 ANTENNAS, 1.5 M DIA., 35 KG EACH
1 RECEIVER, 0.3 X 0.3 X 0.6 M, 50 KG

2 TO 4 KW POWER INPUT

TEST VARIABLES - TRANSMISSION FREQUENCY 1 TO 100 GHZ
- POINTING ACCURACY
- PLASMA ABSORPTION

MEASUREMENTS - PLASMA DENSITY, POWER IN/OUT, POINTING ANGLE

-------------------------------------------------------------------------------------------------------------------------------------

TABLE 1.
SIZING OF TEST BED

<table>
<thead>
<tr>
<th>EXPERIMENT</th>
<th>OPERATING POWER</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>1 KW-</td>
</tr>
<tr>
<td>1. PV-PLASMA</td>
<td></td>
</tr>
<tr>
<td>2. GROUND</td>
<td></td>
</tr>
<tr>
<td>3. CONCENTRATOR</td>
<td></td>
</tr>
<tr>
<td>4. ENERGY WHEEL</td>
<td></td>
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<tr>
<td>5. AMORPHOUS Si</td>
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<tr>
<td>6. NaS BATTERY</td>
<td></td>
</tr>
<tr>
<td>7. Li/TiS2 BATTERY</td>
<td></td>
</tr>
<tr>
<td>8. H2-Br BATTERY</td>
<td></td>
</tr>
<tr>
<td>9. NiH2 BATTERY</td>
<td></td>
</tr>
<tr>
<td>10. HI VOLT PMAD</td>
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</tr>
<tr>
<td>11. MICROWAVE POWER</td>
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</tr>
</tbody>
</table>
### TABLE 2.

**MAJOR COMPONENTS OF TEST BED - PRELIMINARY**

**MECHANICAL** - FIXED PALLET FOR ESS, PMAD, THERMAL, DATA  
- SUN ORIENTED PLATFORM OR ATTACHMENT  
- MSC CAPABILITY  
- ALPHA AND BETA JOINTS FOR POWER/DATA TRANSFER

**ELECTRICAL** - MODULARITY OF POWER CONVERTERS FOR ESS MGMT  
- SOLAR ARRAY VOLTAGE REGULATION UNIT  
- POWER MANAGEMENT AND DISTRIBUTION UNITS  
- SLIP RINGS ACROSS BETA JOINT

**THERMAL** - THERMAL FLUID LOOP ON FIXED PALLET  
- POWER SUPPLY AND CONTROLLER FOR RESISTANCE HEATERS  
- THERMAL BLANKET ATTACHMENTS  
- THERMAL INSTUMENTATION

**DATA/CONTROL** - MICROPROCESSOR FOR DATA, TEST, AND CONTROL  
- DMS INTERFACE

**INSTRUMENTATION** - LANGMUIR PROBE, DENSITY AND PLASMA TEMP  
- SUN SENSOR
APPENDIX 2.

- Three Month Review Summary
## TEST BED SITE TRADE STUDY

<table>
<thead>
<tr>
<th>LOCATION</th>
<th>PRO's</th>
<th>CON's</th>
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<tbody>
<tr>
<td>END OF SOLAR ARRAY BOOM</td>
<td>o SS ALPHA JOINT PROVIDES FULL VIEW OF SUN</td>
<td>o INSTALLATION/REMOVAL REQUIRES EVA</td>
</tr>
<tr>
<td></td>
<td>o DEEP VIEW OF SPACE (THERMAL)</td>
<td>o POSSIBLE ATTITUDE CONTROL IMPACT DUE TO MASS IMBALANCE</td>
</tr>
<tr>
<td>UPPER BOOM</td>
<td>o RMS ACCESS FOR SOME TASKS</td>
<td>o POSSIBLE RCS CONTAMINATION</td>
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<tr>
<td></td>
<td></td>
<td>o HIGH DEMAND REAL ESTATE</td>
</tr>
<tr>
<td>UPPER KEEL/INNER</td>
<td>o RMS ACCESS FOR SOME TASKS</td>
<td>o POSSIBLE RCS CONTAMINATION</td>
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<tr>
<td>SOLAR ARRAY BOOM</td>
<td>o LESS SHADOWING ON SS SOLAR ARRAY THAN UPPER BOOM</td>
<td>o SHADOWS FROM SS APPROXIMATELY 50% OF YEAR</td>
</tr>
<tr>
<td>TWO PART CONFIGURATION PV</td>
<td>o MINIMIZES UPPER BOOM DEMANDS</td>
<td>o COMPLICATES DESIGN</td>
</tr>
<tr>
<td>AND ENERGY STORAGE</td>
<td>o MINIMIZES 'END OF BOOM' COMMANDS ACROSS ALPHA JOINT</td>
<td>o INCREASES COST</td>
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</table>
SAFETY

POTENTIAL HAZARDS                                     SOLUTIONS

HIGH TEMPERATURE RADIATORS                           • REMOTE LOCATION FOR
    - DUMMY LOAD ELEMENTS                              HAZARDOUS HARDWARE
    - THERMAL CONTROL LOOPS                           • EVA PRE-OPERATIONS

HIGH TEMPERATURE EXPERIMENTS                         • WARNING LABELS
    - NaS BATTERIES                                   • PROTECTIVE CONTAINERS

SOLAR ARRAY EXPERIMENTS                              • RECONFIGURATION TO SAFE
    - HIGH VOLTAGE                                   • MODE
    - REVERSE BIASLED CELLS                           • LIMIT SIZE
    - HIGH TEMPERATURE COMPONENTS                    • DELETE EXPERIMENT

PMAD                                                     •
    - HIGH VOLTAGES

OTHER
    - ENERGY WHEEL (HIGH SPEED)
    - Li/TiS2 (EXPLOSIVE)
## TEST BED ENERGY STORAGE EXPERIMENTS

### THERMAL REQUIREMENTS

<table>
<thead>
<tr>
<th>EXPERIMENT</th>
<th>DISCHARGE POWER (WATTS)</th>
<th>DISCHARGE HEAT (WATTS)</th>
<th>RADIATOR TEMP (C)</th>
<th>EXPERIMENT TEMP CONTROL RADIATOR AREA (SQ.M)</th>
<th>ELECTRONICS TEMP CONTROL RADIATOR AREA (SQ.M)</th>
<th>DUMMY LOAD RADIATOR AREA (SQ.M)</th>
<th>TOTAL (SQ.M)</th>
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<tr>
<td>LiT1S2</td>
<td>1000</td>
<td>300</td>
<td>15</td>
<td>.96</td>
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<td>NaS</td>
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<td>NiH2 BIPOLAR</td>
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<td>H2BR2 RFC</td>
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<td>H2O2 RFC</td>
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<td>85</td>
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<td>FLYWHEEL</td>
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<td>480</td>
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<td>1.16</td>
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<td>.60</td>
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# TIME PHASED TEST PLAN

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<td>1. PV-PLASMA</td>
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<td>2. GROUNDING</td>
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<td>3. PV-CONCENTRATOR</td>
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<td>4. ENERGY WHEEL</td>
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<tr>
<td>5. THERMOPHOTOVOLTAIC</td>
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<td>6. BATTERY-NaS</td>
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<td>7. TIME ANNEALING</td>
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<tr>
<td>8. BATTERY-L1/T1S2</td>
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<tr>
<td>9. BATTERY-N1/H2</td>
<td>*****</td>
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<tr>
<td>10. MICROWAVE POWER</td>
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<tr>
<td>11. H2/Br REG. FUEL CELL</td>
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</table>

(1) PRIMARY ARRAY TO BE IN PLACE DURING ABSENCE OF EXPERIMENTAL ARRAYS
(2) CASSIGRAINIAN, SLATS, FRESNEL, NEWTON
(3) TO REMAIN PART OF PRIMARY TEST BED FOLLOWING EXPERIMENTAL PHASE
<table>
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<tbody>
<tr>
<td>1. HI VOLTAGE PMAD</td>
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<td>2. ALBEDO - GBC CELLS</td>
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<tr>
<td>3. DEPLOYMENT/RETRACTION</td>
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<tr>
<td>4. PV-AMORPHOUS S1</td>
<td></td>
<td></td>
<td></td>
<td></td>
<td></td>
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<tr>
<td>5. NEW TECH CELL QUAL.</td>
<td>***</td>
<td></td>
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<td>***</td>
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<tr>
<td>6. REAL TIME CELL DEG.</td>
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<td>7. ON ORBIT ARRAY REPAIR</td>
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<td>8. STANDARDS CALIBRATION</td>
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APPENDIX 3.

- Detailed Conceptual Equipment Design
COMPARISON OF RESOURCES ON TEST BED AND FREE FLYER

<table>
<thead>
<tr>
<th></th>
<th>SPACE STATION (SS)</th>
<th>FREE FLYER</th>
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<tbody>
<tr>
<td>COMMAND/TELEMETRY</td>
<td>AVAILABLE</td>
<td>AVAILABLE</td>
</tr>
<tr>
<td>ELECTRICAL POWER</td>
<td>AVAILABLE</td>
<td>MORE LIMITED THAN SS</td>
</tr>
<tr>
<td>SOLAR ILLUMINATION AREA</td>
<td>AVAILABLE</td>
<td>MORE LIMITED THAN SS</td>
</tr>
<tr>
<td>THERMAL</td>
<td>AVAILABLE IF NEEDED</td>
<td></td>
</tr>
<tr>
<td>EVA OR RMS ACCESSABILITY</td>
<td>AVAILABLE SCHEDULING FLEXIBLE</td>
<td>AVAILABLE AT RETRIEVAL</td>
</tr>
</tbody>
</table>

* ASSESSMENT REQUIRES CONSIDERATION OF DATA BASE ON ALOCATION OF RESOURCE.

** COMPARISON REQUIRES QUANTIFICATION, AND COMPARISON WITH EXPERIMENT REQUIREMENTS.
END OF SOLAR ARRAY BOOM
SIZE COMPARISON WITH SPACE STATION

ADVANCED POWER SYSTEM TEST BED (APSTB)

APSTB LOCATED ON SPACE STATION

Ford Aerospace & Communications Corporation
APPENDIX 4.

- Preliminary Evolutionary Plan
LEVEL 1. BASIC PRINCIPLES OBSERVED AND REPORTED.

LEVEL 2. CONCEPTUAL DESIGN FORMULATED.

LEVEL 3. CONCEPTUAL DESIGN TESTED ANALYTICALLY OR EXPERIMENTALLY.

LEVEL 4. CRITICAL FUNCTION BREADBOARD DEMONSTRATION.

LEVEL 5. COMPONENT OR BRASSBOARD MODEL TESTED IN RELEVANT ENVIRONMENT.

LEVEL 6. COMPONENT OR ENGINEERING MODEL TESTED IN RELATIVE ENVIRONMENT.

LEVEL 7. ENGINEERING MODEL TESTED IN SPACE.

LEVEL 8. BASELINED INTO PRODUCTION DESIGN.
<table>
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<tr>
<th>Experiment/Measurements</th>
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<th>Tasks To Do</th>
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<tbody>
<tr>
<td>1. Plasma Interaction</td>
<td>2</td>
<td>- Shuttle flight o VOLT o Interaction with S/A Segments o Test Bed o Planning based upon Shuttle flight</td>
<td>LMSC *Sharp</td>
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<td>- High voltage Plasma</td>
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<td>2. High Voltage PMAD</td>
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<td>- Plasma Interaction</td>
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<td>- System efficiency</td>
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<td>- System safety</td>
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<td>3. Solar Array -</td>
<td>4</td>
<td>- Ground Test EM - Shuttle Flight o Deployment o Pointing Accuracy o Power capability o Protective Coatings</td>
<td>GE TRW *Aerospatiale</td>
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<tr>
<td>Concentrator</td>
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<td>- Test Bed o Protective Coatings o Long term performance</td>
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<td>- Verify Technology in</td>
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<tr>
<td>Space</td>
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<td>- Deployment</td>
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<td>- Pointing Accuracy</td>
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<td>- Power capability</td>
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<td>- Protective Coatings</td>
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<td>4. Energy Wheel</td>
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<tr>
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<td>- Power capability</td>
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<td>- Safety</td>
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* Inputs Received From Letter Survey
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<tr>
<td>5. Amorphous Silicon Array</td>
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<td>- Degradation</td>
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<td>o Stow/Deploy</td>
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<td>- Power Output</td>
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<td>o Stow/Deploy</td>
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<td></td>
<td>o Efficiency</td>
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<td>6. NaS Battery</td>
<td>5</td>
<td>- Shuttle Flight</td>
<td>*USAF</td>
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<tr>
<td>- Zero-G Operation</td>
<td></td>
<td>o Not Feasible</td>
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<tr>
<td>o Wicking in both electrodes</td>
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<td>- Test Bed</td>
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<td>- Launch</td>
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<td>o Launch Inactive, Activate on Test Bed</td>
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<td>o electrolyte breakage</td>
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<td>o Radiator Surface</td>
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<td>7. Li/TiS Battery</td>
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<tr>
<td>o Zero Gravity (Long Term)</td>
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<td>o Zero G (Short Term)</td>
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<td>o Launch</td>
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<td>- Test Bed</td>
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<td>o Zero G (Long Term)</td>
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<td>9. NiH2 Bipolar Battery</td>
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<tr>
<td>12. Thermophotovoltaics</td>
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<td>- Surface properties</td>
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<td>- Qualification of solar cell technology</td>
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<td>- Vaporization of high temperature surfaces</td>
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<td>- Deposition of materials onto reflective surfaces</td>
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<td>- Atomic Oxygen Interaction</td>
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<td>- Solar Cell Technology</td>
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<tr>
<td>- Cell design for specific spectral output</td>
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<td>- Inputs Received From Letter Survey</td>
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</tbody>
</table>
## INDUSTRY CONTACTS

1. **LITHIUM BATTERIES**  
   MR. E.U. COSTOGUE  
   JET PROPULSION LABORATORY  
   SPACE POWER SECTION  
   4800 OAK GROVE DRIVE  
   PASADENA, CALIF. 91109

2. **AMORPHOUS SILICON**  
   DR. JOSEPH HANAK  
   SOVONICS  
   6180 COCHRAN ROAD  
   P.O. BOX 39608  
   SOLON, OHIO 44139

3. **CONCENTRATOR ARRAYS**  
   MR. LIONEL PELENC  
   AEROSPATIALE  
   100 BOULEVARD DU MIDE BP99  
   CANNES, FRANCE 06322

4. **SODIUM SULPHER BATTERIES**  
   LT. ROSS DUEBER  
   UNITED STATES AIR FORCE  
   WRIGHT PATTERSON AIR FORCE BASE  
   POWER TECHNOLOGY BRANCH  
   OHIO 45433

5. **PLASMA TESTING**  
   MR. AKIO SUZUKI  
   DIVISION GENERAL MANAGER  
   SHARP ELECTRONICS  
   2613-1 ICHINOMOTO, TENRI-CITI  
   NARA 632, JAPAN

6. **NICKEL HYDROGEN BIPOLAR BATTERIES**  
   MR. CHARLES KOEHLER  
   FORD AEROSPACE AND COMMUNICATIONS CORPORATION  
   3939 FABIAN WAY M/S G-45  
   PALO ALTO, CALIF. 94303
# Test Bed Resource Requirement Planning

## Energy Storage Experiments

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| TOTAL REQUIREMENTS (Watts) | 880 | 1650 | 2530 | 1650 | 880 | 1230 | 1540 |

Maximum power scheduled: 2530 Watts