Some Features of Surface Pressure Fluctuations in Turbulent Boundary Layers With Zero and Favorable Pressure Gradients

B. E. McGrath and R. L. Simpson

Virginia Polytechnic Institute and State University
Blacksburg, Virginia

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SUMMARY

Various researchers are interested in the structure of the surface pressure fluctuations for the development and use of noise prediction techniques for helicopter and turbomachinery rotors. This study, conducted in the Virginia Tech low speed boundary layer wind tunnel, covered the effects of zero and favorable streamwise pressure gradient flows on the surface pressure fluctuation spectra, coherence and convective wave speeds in turbulent boundary layers for momentum Reynolds numbers from 3000 to 18,800. The acceleration parameter, $K$ is near $2 \times 10^{-7}$ for the favorable pressure gradient flow. Small pinhole condenser microphones were used to obtain the surface pressure fluctuation data for all test cases. The longitudinal and lateral coherence functions and the convective wave speeds were obtained for both streamwise pressure gradient flows.

The results presented are for the surface pressure fluctuation spectra nondimensionalized by different groupings of the outer and inner boundary layer variables. The grouping using the outer variables, $U_e$, $\tau_w$ and $\delta_1$ collapse the spectra for the low to middle range of frequencies for most test cases. The grouping using the inner variables, $U_t$ and $\nu$, collapse the spectra for the middle to high range of frequencies for all test cases. The
value of $p'/t_w$ was near 3.8 and 2.8 for the smallest values of $d^+$ in the zero and favorable pressure gradient flows, respectively.

The spectral data was corrected using the correction developed by G.M. Corcos, but the pinhole correction developed by Bull and Thomas was not used in the data reduction process. However, some discussion is included on the effects of the pinhole correction for the results of this study.

The coherence exhibits a decay that is not exponential in some cases, but the Corcos similarity parameters $\omega\Delta x/U_c$ and $\omega\Delta z/U_c$ collapse the data for all test cases. The ratio of $U_c/U_e$ increases with $\omega\delta_1/U_e$ up to $\omega\delta_1/U_e$ on the order of unity, where $U_c/U_e$ becomes nearly constant. This was observed in the present results for both streamwise pressure gradient flows.

The experimental results presented show good agreement with previous research.
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LIST OF SYMBOLS

c ............... speed of sound, m/s
C_f ............... skin friction coefficient
d ............... diameter of microphone, m
d^+ ............... nondimensional diameter, dU_t/ν
G_{xx} ............... the power spectrum at position x
G_{zz} ............... the power spectrum at position z
G_{zx} ............... the cross power spectrum between position z
                    and position x
H ............... shape factor, δ_l/θ
I_n ............... normalized quadrature of the cross spectrum
                    for frequency n
k ............... wavenumber, ω/U_c
K ............... acceleration parameter for nonzero
                    streamwise pressure gradient flows,
                    \((ν/U_e^2)(dU_e/dx)\)
K_1,K_3 ........... exponential decay constants for longitudinal
                    and lateral directions
n ............... frequency, Hz
P ............... instantaneous pressure, Pa
P_o,P ............... mean and fluctuating wall pressures, Pa
P' ............... rms wall pressure fluctuation, Pa
q_e ............... local free stream dynamic pressure, Pa
r ............... transducer radius, m
\( R_n \) ............ normalized co-spectrum of the cross spectrum for frequency \( n \)

\( Re \) ............. Reynolds number

\( Re_{\delta_1} \) ........ displacement thickness Reynolds number

\( Re_{\delta} \) ........ momentum thickness Reynolds number

\( U_c \) ............. convective wave speed, m/s

\( U_e \) ............. local free stream velocity, m/s

\( U_i \) ............. mean velocities in \( i \) directions, m/s

\( u_i \) ............. fluctuating velocities in the \( i \) direction, m/s

\( U_f \) ............. friction velocity, m/s

\(-uv\) ............ kinematic Reynolds shear stress, \((m/s)^2\)

\( V_i \) ............. instantaneous velocity in the \( i \) directions, m/s

\( V \) ............. volume, \( m^3 \)

\( x, y, z \) ........ streamwise, normal to wall, spanwise positions, m

\( \Delta x, \Delta y, \Delta z \) .... spatial separations in the respective directions, m

**GREEK SYMBOLS**

\( \alpha \) ............. ratio of streamwise length scale to length scales in other directions

\( \gamma^2 \) ............. coherence function, \( \gamma^2 = |G_{xx}|^2 / G_{xx} G_{zz} \)

\( \gamma \) ............. square root of the coherence function
\[ \delta \quad \text{boundary layer thickness} \]
\[ \delta_1 \quad \text{boundary layer displacement thickness} \]
\[ \theta \quad \text{boundary layer momentum thickness} \]
\[ \mu \quad \text{coefficient of viscosity, kg/m}\cdot\text{sec} \]
\[ \nu \quad \text{kinematic viscosity, m}^2/\text{sec} \]
\[ \rho \quad \text{density, kg/m}^3 \]
\[ \phi \quad \text{phase angle, } \omega A x/U_c \text{ and } \omega A z/U_c \]
\[ \hat{\phi}(\omega) \quad \text{power spectrum of fluctuating pressure as a function of } \omega, \text{ where } p^2 = \int_{-\infty}^{+\infty} \hat{\phi}(\omega) d\omega \]
\[ \hat{\phi}(k) \quad \text{power spectrum of fluctuating pressure as a function of } k, \text{ where } p^2 = \int_{-\infty}^{+\infty} \hat{\phi}(k) dk \]
\[ \omega \quad \text{radial frequency, } 2\pi n \text{ radians} \]

**SUBSCRIPTS**

\[ a \quad \text{acoustic contribution to fluctuating pressure} \]
\[ t \quad \text{turbulent contribution to fluctuating pressure} \]
\[ n \quad \text{frequency, Hz} \]
\[ 1,2 \quad \text{microphone unit or position number} \]
The study of surface pressure fluctuations in a turbulent boundary layer flow has been of interest to researchers for many years. Surface pressure fluctuations that occur in turbulent flow are noise sources. Helicopter and turbomachinery rotors, aircraft and ships are examples of practical devices whose surface turbulent boundary layers generate pressure fluctuations that contribute to the generation of noise. Designers and researchers are most interested in methods for predicting and reducing flow noise due to pressure fluctuations. Brooks and Schlinker (1982) give a review on the recent progress in rotor noise research.

The problem faced by many researchers and designers is the lack of detailed information on the relationship between the turbulent flow field and the resulting pressure fluctuations. A recent effort by Brooks and Hodgson (1981) shows the development of a noise prediction method for turbulent boundary layer flow. Brooks and Hodgson used a NACA 0012 airfoil in their experimental study to relate the turbulent flow field to the surface pressure spectra, cross spectra and convective wave speeds. Thus, the prediction method uses a statistical model of the turbulent boundary layer pressure field and empirical relations of the convective wave speeds. Using this information the cross
spectra of the flow field are predicted and the resultant rms pressure fluctuation can be calculated, therefore giving an overall estimate of the resultant noise. These recent studies have given some direction to researchers and designers interested in the study of flow noise due to pressure fluctuations.

The experiments performed in Virginia Tech's boundary layer wind tunnel are an effort to obtain detailed velocity and surface pressure experimental data for two zero and one favorable streamwise pressure gradient turbulent boundary layers. Some previous research on these types of flow has been conducted by Bradshaw (1967), Willmarth (1975), Schloemer (1967), Blake (1970), Bull (1967) and others that are included in the list of references.

The wind tunnel and the test flows are discussed in section 3. A flat plate 8 meters in length and 0.9 meters in width was used in the present experiments. A cross section of the wind tunnel test section is shown in Figure 1. Zero pressure gradient flows between Reₘ of 3300 to 18,800 were examined. The favorable pressure gradient flow permitted examination of the streamwise flow properties between Reₘ of 3000 to 9000. The favorable pressure gradient flow work is an effort to provide information where little has been previously provided. The fluid dynamic properties of these flows were obtained using a hot-wire anemometer. The surface pressure spectra and convective wave speeds were
measured using miniature pinhole condenser microphones mounted flush with the surface as discussed in section 4.1. Two sets of microphones were located at each x-location and separated in the spanwise direction by approximately one-third of a meter. Because streamwise acoustical and unsteady waves are the same at both spanwise locations at any instant, the spectrum of the difference of these time-varying signals is related only to the turbulent surface pressure spectrum, as discussed in section 4.2 below. This setup permitted measurement of turbulence generated pressure spectral data in a tunnel that is not acoustically quiet. This method of data acquisition provides an advantage over previous work because no additional work was needed to quiet the flow to measure the pressure fluctuations due only to the flow field fluctuations.

The results that are presented in section 5 are the power spectra of the surface pressure fluctuations, rms pressure values, some flow field properties, convective wave speeds and the square root of the coherence in the streamwise and spanwise directions. The results of previous work are compared in section 6 to the results obtained in this study. Some development of the relationships among the flow field, pressure spectra, coherence spectral magnitude and convective wave speeds is also included.
2.1 FORMULATION

Consider an incompressible turbulent flow, which is the case for the present experiments. The Navier-Stokes equations define the relationship between the fluctuating pressures and fluctuating velocities. In vector form the equation is written as follows.

\[
\frac{\partial \bar{V}}{\partial t} + \bar{V} \cdot \nabla \bar{V} + \nabla P/\rho = \nu \nabla^2 \bar{V} \quad (1)
\]

\(\bar{V}\) is the velocity vector, \(\rho\) is the density, \(\nu\) is the kinematic viscosity and \(P\) is the pressure. For incompressible flow \(\rho\) is constant and we also assume that \(\nu\) is constant. Taking the divergence of each term in the equation above and making use of the continuity equation

\[
\nabla \cdot \bar{V} = 0 \quad (2)
\]

we obtain

\[
\frac{\partial^2 P}{\partial x_i^2} = -\rho q \quad (3)
\]

where \(q\) is given by the following
For turbulent flow we can define the velocity vector as

\[ \mathbf{V}_i(\mathbf{x}, t) = \mathbf{U}_i(\mathbf{x}) + \mathbf{u}_i(\mathbf{x}, t) \] (5)

and the pressure as

\[ P(\mathbf{x}, t) = P_o(\mathbf{x}) + p(\mathbf{x}, t). \] (6)

Now placing these terms into equation (3) and rearranging we obtain Poisson's differential equations for the pressure fluctuations,

\[ \frac{\partial^2 p}{\partial x_i^2} = -2\rho \left( \frac{\partial U_j}{\partial x_i} \right) \left( \frac{\partial u_j}{\partial x_i} \right) - \rho \frac{\partial^2 (u_i u_j - u_i u_j)}{\partial x_i \partial x_j}. \] (7)

\( U_i \) and \( u_i \) are the mean and fluctuating velocities in the \( x_i \) direction. The first term of the RHS of this equation represents the turbulence-mean shear interaction and the second term represents the turbulence-turbulence interaction. To obtain a solution to equation (7) for surface pressure fluctuations, we integrate the equation for a wall-bounded flow. Neglecting the contribution of the
surface integrals, then the fluctuating pressure at some point \( \overline{X} \) on the wall is given by

\[
p(\overline{X}) = \rho/2\pi \int_{Y>0} q(\overline{X}_s) dV(\overline{X}_s)/|\overline{X} - \overline{X}_s| \quad (8)
\]

where the volume integration is at all positions \( \overline{X}_s \) over the entire half-space containing the flow. This equation shows that the surface pressure fluctuations are produced from sources in a large region of the flow, but contributions from various sources drop off rapidly with increasing distance from the point \( \overline{X} \) under consideration.

Several attempts have been made to obtain the surface pressure field theoretically from equation (8), but Willmarth (1975) has pointed out that such efforts suffer from the lack of accurate information about the fluctuating velocity field in the turbulent boundary layer flow. Thus, such efforts need confirmation by experimental data and this experimental investigation is an effort to provide data to bridge that gap. Some earlier research that deals with the theoretical and experimental aspects of surface pressure fluctuations are discussed in the following sections.

2.2 CALCULATIONS OF PANTON AND LINEBARGER

Panton and Linebarger (1974) developed a numerical solution for the wall pressure spectra in two-dimensional
turbulent boundary layers. Their solution were for zero and adverse pressure gradient equilibrium boundary layers. The results seem to describe the essential features observed in experiments. They used Coles' laws of the wall and wake for the mean velocity profiles. A scale-anisotropic model of the spatial correlations of \( v \) was used together with the assumption that \( v \) is proportional to \( \sqrt{-uv} \). Only the turbulence-mean shear interaction term in equation (7) was modeled since the turbulence-turbulence interaction contributes a small portion to the mean-square value in such flows.

Their spectral results show larger contributions at higher \( \text{Re} \) for \( k\delta<20 \), than for some of the previous experimental studies. Contributions at these low frequencies are due to the outer region velocity and turbulence structure and depend on the pressure gradient. An overlap region between the low frequency outer-flow-dominated part and the high frequency near-wall viscous-sublayer-dominated part of the spectrum varies with \( k^{-1} \) as observed by Bradshaw (1967). Their calculation results are approximated by

\[
kF(k)/\tau_w^2 = 1.73a^{0.9}, \quad \text{for } k\nu/\bar{U}_t < 0.06.
\]

Here \( a \) is the ratio of the streamwise length scale to length scales in other directions, which influences the spatial
correlation of $v$. For higher frequencies, the spectral variation is given by

$$kF(k)/\tau_w^2 = 0.0173(k\nu/U_t)^{-2}, \text{ for } k\nu/U_t \geq 0.1. \quad (10)$$

Both of these equations are independent of $Re$ and are scaled on the wall shear stress.

Because the low frequency part of the spectrum is $Re$ dependent, the mean square pressure fluctuation increases with $Re$. The equation

$$p'/\tau_w^2 = 0.52\alpha^{0.9}(\ln|U_\delta/\nu| + 9.24) \quad (11)$$

fits Panton and Linebarger's calculations for a zero pressure gradient with $\alpha=1,2$ and $3$ with Coles' wake parameter $\Xi=0.6$ where

$$U_\delta/\nu = K(U_\delta/\nu-65)/(1+\Xi) \quad (12)$$

Figure 2 shows the results from equations (11) and (12). Panton and Linebarger show that $p'/\tau_w$ varies between 2.9 and 3.1 for $4000 \leq Re_\theta \leq 40,000$.

Panton and Linebarger also include some calculations of the convective wave speeds for zero pressure gradient flows. Their results show that the wave speed decreases with increasing $n$ or $k$ and increases with increasing $Re$. The
calculations don't include the cross spectral density or coherence functions. Therefore, comparisons here are restricted to the surface pressure spectra and wave speeds for zero pressure gradient flows.

2.3 SOME PREVIOUS EXPERIMENTAL SURFACE PRESSURE FLUCTUATION STUDIES

Researchers have studied pressure fluctuations in different streamwise pressure gradient flows using different pressure transducers of various sizes. Thus far most studies have been in fair agreement with one another. Corcos (1963) revealed that there is attenuation of the pressure fluctuations at frequencies where the wavelength is of the same order of magnitude or smaller than the diameter, 2r, of the pressure transducer diaphragm. When the length scales of the pressure fluctuations are small, there is an averaging of the amplitudes over the surface of the transducer. Thus, some attenuation occurs at the higher frequencies. Corcos (1963,1967) proposed that a correction be applied to the spectra as a function of \( \omega r/U_c \), where \( U_c \) is the convective wave speed. The correction amplifies the spectrum by as much as 3 dB for the higher frequencies. Most researchers agree that the attenuation occurs at high frequencies and one must use the correction proposed by Corcos to correct the spectral results as was done here, Schewe (1982) indicated that the
Corcos correction is not large enough when $\omega r/U_c > 4.0$. However, Schewe did not suggest how much larger the correction should be.

Bull and Thomas (1976) performed a study in zero pressure gradient flows using two different transducer mountings. One was a pinhole piezoelectric transducer with the diaphragm recessed from the surface and the other was a piezoelectric transducer mounted flush with the surface. The pinhole transducer caused a small discontinuity on the surface while the flush mounted piezoelectric transducer kept the surface smooth and continuous. The results from Bull and Thomas (1976) show that there is a large difference between transducers. The study then indicated that there was an increase in spectral density for the pinhole transducer for nondimensional frequencies of $0.1 \leq \omega \nu / U_t^2 \leq 2.0$. At these frequencies, the wavelengths of the surface pressure fluctuations are on the order of and smaller than the pinhole. The ratio of spectral densities $\phi(\omega)_p / \phi(\omega)_x$ can be as large as 3.5 to 4.0, where the subscript $p$ denotes the pinhole results and the subscript $x$ denotes the flush surface results. Bull and Thomas contend that there is a rather large effect due to interaction of the turbulent boundary layer with the small pinhole. This effect is referred to here as the Bull and Thomas effect. A correction for the spectrum was provided in their paper. However, the correction was not accounted for in the results shown here.
Little explanation of the effect due to the pinhole was provided by Bull and Thomas and was not found elsewhere in the literature.

2.3.1 ZERO PRESSURE GRADIENT

Table 1 gives an overview of earlier experiments performed in zero pressure gradient flows. Various ranges of Re₀ and transducer size were used. The transducer size is perhaps the most important consideration one must look at before comparing works. Figure 3 shows the rms pressure fluctuation nondimensionalized on qₑ versus d⁺, where d⁺ is the nondimensional transducer diameter. The diameter is nondimensionalized on the inner variables, showing some importance on the turbulence-mean shear interaction. The plot shows a decrease in p'/qₑ with increasing d⁺, where at a certain point p'/qₑ becomes constant regardless of d⁺. As mentioned previously the resolution of the high frequencies is important, thus the microphone diameter needs to be small to reduce the value of d⁺. To reduce the value of d⁺ some researchers have used very small sensing diameters, obtained by using a pinhole atop the transducer diaphragm. Blake (1970) and Dinkelacker and Langeheineken (1982) used a pinhole type microphone. Others have used flush mounted piezoelectric or condenser microphones.
Now with the consideration of the size effect, we can make some observations about previous experiments. Figure 2 shows rms pressure nondimensionalized on the wall shear stress. This parameter also shows the turbulence-mean shear interaction which is the dominant feature in wall bounded flows. From the figure \(p'/\tau_w\) varies between 1.8 and 3.8. Blake (1970) shows that \(p'/\tau_w\) is approximately 3.6. Blake used pinhole microphones, but as shown in Figure 3 the agreement with other researchers who did not use pinhole microphones is very good. Panton and Linebarger's calculation show \(p'/\tau_w\) between 2.9 and 3.1.

Most other researchers are below Panton and Linebarger's calculations. Bull and Thomas (1976) show \(p'/\tau_w\) to be near 2.8, however, as seen in Figure 3 their values of \(d^+\) are nearly the same for Blake (1970) but Bull and Thomas used flush mounted piezoelectric transducers. Schloemer had a fairly large value of \(d^+\), and gives the lowest values of \(p'/\tau_w=1.63\). As discussed in Lim (1971), values of \(p'/\tau_w\) are predicted to range anywhere from 2.56 to 6. However, in a personal conversation with Lim (1971), Hodgson estimates that \(p'/\tau_w>4\). Under these considerations, one is led to believe that the larger values of \(p'/\tau_w\) and \(p'/q_w\) are correct for smaller \(d^+\).

Table 1 also gives the spectral level for various researchers at a value of the nondimensional frequency, \(\omega^*_1/U_e=1.0\). We see that the spectral level is \(-51\pm1.5\) dB,
where \( dB = 10 \log_{10} |\psi(\omega)U_e/\sigma_e^2|_1 \). This value sets the spectral level because at \( \omega \delta_1/U_e = 1 \) the spectrum is not influenced by the high frequency resolution limitations or the uncertainty of the lowest frequencies. The agreement here is very good. Spectral trends for \( \omega \delta_1/U_e \geq 1.0 \) show that for most experiments the spectrum varies like \( n^{-1} \), especially in flows at larger \( \text{Re}_\theta \) values. The spectra tend to drop off much faster above \( \omega \delta_1/U_e > 5 \), where the spectrum varies between \( n^{-4} \) and \( n^{-6} \). Figure 4 shows the mean spectral data for several researchers. Fairly uniform trends are seen in the previous works and are nearly independent of transducer type, but are dependent on transducer size.

Cross spectral measurements in zero pressure gradient flows were obtained by Schloemer (1967), Bull (1967), Blake (1970), and Corcos (1964). Corcos (1963) and Brooks and Hodgson (1981) propose that the cross spectrum in either the lateral or longitudinal direction decay exponentially with the phase angle, \( \phi \), as shown in the equation

\[
C_{zx} = \psi(\omega) \exp(-K_3 \omega \delta z/U_c - K_1 \omega \Delta x/U_c + iK_3 \Delta x). \tag{13}
\]

In this equation \( K_1 \) and \( K_3 \) are the decay constants for the cross spectrum and the square root of the coherence, \( \gamma \). Equation (13) also indicates that \( \gamma = e^{-K\phi} \).

For two different \( \text{Re}_\theta \), Brooks and Hodgson show that \( K_1 = 0.19 \) and \( 0.14 \) for \( \Delta x/\delta > 3 \) and \( K_3 = 0.62 \) and \( 0.58 \),
respectively. For $\Delta x/\delta_1<3$, $K_1$ was a stronger function of $\Delta x/\delta_1$, with $K_1$ values as large as 0.23. Although Brooks and Hodgson's study was on a weak adverse pressure gradient flow, the approximately exponential decay is present for both the zero and favorable pressure gradients as well. In the zero pressure gradient case, Bull (1967) and Blake (1970) show good agreement with $K_1=0.1$ and $K_3=0.54$ for $\Delta x/\delta_1>3$, for smaller $\Delta x/\delta_1$, Bull found $K_1$ to be as large as 0.15. These past experiments show that for longitudinal spacings the decay of the cross spectra is small. This says that the pressure fluctuations convecting downstream remain coherent for large distances.

The square root of the coherence in the lateral direction decays much faster than for the longitudinal decay as observed in all previous work. As seen in Table 1 for $\Delta x/\delta_1>3$, the values of $K_3$ are at least 5 times greater than $K_1$. This indicates that the pressure fluctuations are not as coherent over the spanwise direction as in the streamwise direction.

Also, the convection velocities at which these fluctuations travel increase with increasing frequency and at high frequencies remain nearly constant at a value between 70 and 80 percent of the free-stream velocity, as shown by Schloemer (1967), Blake (1970), Bull (1967). Schloemer's data show that there is an apparent increase in convection velocity with increased transducer spacing.
2.3.2 FAVORABLE PRESSURE GRADIENT

There are fewer studies of surface pressure fluctuations in accelerating flows than for zero pressure gradient flows. Schloemer (1967), Burton (1973) and Schewe (1983) have performed the bulk of the work for favorable pressure gradient flows and a summary of the results is given in Table 2. As discussed previously, the transducer resolution and size is even more important in accelerating flows because the viscous region is much smaller than in the zero or adverse pressure gradient. This means that for small transducers, the nondimensional diameter, $d^+$ is larger for the same transducer in a zero or adverse pressure gradient flow. Therefore, one must closely examine the data for resolution and transducer size. None of these previous researchers have used a pinhole transducer in a favorable pressure gradient flow. Figure 3 shows that, for the favorable pressure gradient case as well, the value of $p'/q_e$ increases with decreasing $d^+$. This is not surprising since from previous discussion we know that the resolution increases with decreasing transducer diaphragm size. The data for the favorable pressure gradient flow follows a similar trend as seen for the zero pressure gradient case. Figure 5 shows $p'/\tau_w$ verses displacement thickness Reynolds number. Burton (1973) shows $p'/\tau_w$ is near 2 for several Reynolds numbers, but the value of $d^+$ is relatively large. Schloemer (1967)
gives $p'/\tau_w$ near 1 and has a very large $d^+$. Schewe (1982) gives values of $p'/\tau_w$ between 2.48 and 1.2 for various $d^+$. Schewe's and Schloemer's data for the larger $d^+$ suffer from poor transducer resolution and we can conclude that the data for the smaller $d^+$ are more correct. Bull's (1967) slightly accelerating flow shows $p'/\tau_w$ to be 2.1 to 2.8 for relatively large $d^+$. Burton's, Schloemer's, and Schewe's data agree well for similar values of $d^+$, adding more confidence to the fact that Schewe's smallest $d^+$ gives the most reasonable value.

Comparing the spectral levels at $\omega \delta /U_e = 1$, we see that $10\log_{10} |\phi(\omega)| U_e/\delta^2$ is approximately $-49 \pm 2.5$ dB. The spectral level for the favorable pressure gradient flow is slightly higher than for the zero pressure gradient flow. The region of $n^{-1}$, spectral variation seems to be present for most of the previous work, however, the region spans over a smaller variation of $\omega \delta /U_e$ than for the zero pressure gradient flow. At the higher frequencies, the spectral level varies much like the zero pressure gradient, but the frequency at which the drop off occurs is lower than for the zero pressure gradient flow. Thus, there is not as much energy at the highest frequencies in the favorable pressure gradient flow. Figure 4 shows a plot of the mean spectra from several researchers.

Cross spectral measurements in accelerating flows were performed by Burton and Schloemer. The square root of the
coherence shows an approximately exponential decay in both longitudinal and lateral directions. The longitudinal decay is given by the constant $K_1$ and we see that both Burton and Schloemer show $K_1 \approx 0.1$ for $\Delta x/\delta_1 \geq 3$ or so. This is a slightly slower streamwise decay as compared to the zero pressure gradient case. The decay for the lateral direction is given by $K_3 \approx 0.4$ which is again a slower decay compared to the zero pressure gradient flow. The reason for the slower decay is the fact that in an accelerating flow the flow is self-similar and more coherent over much greater streamwise and spanwise directions. This leads to the pressure fluctuations being coherent for longer distances in both directions.

The convection velocities of these fluctuations are shown to increase with increasing frequency and become nearly constant at higher frequencies. When the convection velocity becomes constant, it remains at a value between 50 to 60 percent of the free stream velocity. Burton and Schloemer both show this trend. The constant value of the convection velocity is about 10 percent lower in a favorable pressure gradient flow than a zero pressure gradient flow. Schloemer also shows that the wave speeds are a function of the transducer spacing, and Brooks and Hodgson show this for an adverse pressure gradient. Physically, this trend is hard to believe since the spacing can have no direct effect on the flow. However, we can say that the more coherent large-scale
structures contribute more to the apparent convection speed at increasing spacing, which tends to make the convection velocity appear to be a function of the transducer spacing.
3.0 DESCRIPTION OF THE WIND TUNNEL AND TEST FLOWS

3.1 WIND TUNNEL

The wind tunnel used at Virginia Tech is the same facility used in previous work at Southern Methodist University (Simpson, et al., 1981; Shiloh et al., 1981; and Simpson et al., 1983). The mainstream flow of the blown open-circuit wind tunnel is introduced into the test section after passing through an air filter, air chiller, blower, fixed-setting flow damper, a plenum, seven screens for removal of some free stream turbulence and finally through a four to one contraction ratio nozzle to accelerate the flow to test speed and to remove additional free-stream turbulence intensity.

Figure 1 is a side view of the eight meter long and 0.91 meter wide test section. The side walls are made of plate float glass, while the upper wall is plexiglas. The zero and favorable pressure gradient flows are obtained by placing sections of plywood inside the test section and supporting the 'false upper wall' from above. The supports allowed for adjustments to the wall to obtain the desired contour. Figure 1 shows the wall contour for both flows. The solid and dashed lines are the contours for the zero and favorable streamwise pressure gradients, respectively. The corner gaps
between the false wall and glass side walls were covered with a flexible polyurethane plastic sheet for preventing flow leakage at these corners. The boundary layer along the test section was turbulent. In order to insure turbulent flow, a 6 mm forward facing step at the leading edge of the test wall for the test section was used to trip the boundary layer.

3.2 TEST FLOWS

Measurements of test flow velocities were obtained by Ahn (1986) using a single channel hot-wire anemometer. For all measurements the temperature was 25±0.5 °C and $v=1.56 \times 10^{-5}$ m$^2$/s. Two different zero pressure gradient flaws were used to obtain data. In both of these flows the flow accelerated for the first 1.6 m of the test section. All measurements of the zero pressure gradient flows were obtained at downstream streamwise stations (Table 3). The higher speed flow for the zero pressure gradient was used to obtain data at high values of $Re_b$. Both flows are not exactly zero pressure gradient flow since the free-stream velocity is nearly constant but has a ±0.3 m/s variation. Table 4 shows that the acceleration parameter $K = (v/U_e^2)dU_e/dx$ is about $2 \times 10^{-7}$ over most of the measured length of the favorable pressure gradient flow.

Tables 3 and 4 give the boundary layer properties for all test flows derived from measured velocity profiles. For
the zero pressure gradient flows, the boundary layer profiles were measured at several streamwise locations. For the streamwise locations where hot-wire measurements were not obtained we can linearly interpolate to get the desired boundary layer quantities. In the favorable pressure gradient case it is not as simple to obtain these values. We must use a momentum integral technique to calculate the boundary layer properties at x-locations not measured with the hot-wire. The equation used to calculate the boundary layer properties is the following,

\[ \frac{\theta(x)}{\nu} = \frac{\theta(x_0)}{\nu} \left( \frac{U_e}{U_e(x)} \right)^{4.11/4} \left( \frac{U_e}{U_e(x_0)} \right)^{4.11/4} = 0.016 \int_x^{x_0} \left( \frac{U_e}{U_e(x)} \right)^{3.86} \, dx. \] (14)

This equation is derived from the momentum integral equation, the Ludwieg-Tillmann skin friction equation, and an assumed constant shape factor, \( H = 1.29 \) (Kays and Crawford, 1980). The skin friction coefficient was obtained from the slope of the semi-logarithmic velocity profile region in clauser plots, which is very close to the skin friction coefficient calculated from the Ludwieg-Tillmann equation (Ahn, 1986)

\[ C_f = (0.246 \times 10^{-0.678H}) (U_e \theta/\nu)^{-0.268}. \] (15)

Results for \( \sqrt{C_f^2 = U_t/U_e} \) from the Ludwieg-Tillmann equation are presented in the tables.
For the zero pressure gradient cases, the boundary layer properties agree with previous studies and give us confidence that the boundary layers are normal two-dimensional boundary layers (Ahn, 1986). The trends show that $\delta$ and $\delta_1$ increase nearly proportional to $x^{0.8}$ while the skin friction coefficient decreases nearly proportional to $x^{-0.2}$.

For the favorable pressure gradient case, the boundary layer properties also indicate a good two-dimensional turbulent boundary layer. The trends also show an increase in $\delta$ and $\delta_1$ with increasing $x$, up to a point where the streamwise pressure gradient causes $\delta$ and $\delta_1$ to decrease with increasing $x$. This streamwise variation of $\delta_1$ is also predicted by equation (14) and is discussed in more length by Ahn (1986). The skin friction coefficient shows a decrease and then an increase with $x$. $Re_\theta$, $U_+$, and $f_W$ all show an increase in increasing $x$. The internal self-consistency of the data provides additional confidence in the quality of this experimental flow (Ahn, 1986).

$U^+$ versus $y^+$ velocity profiles near the wall are in good agreement with earlier studies. The semi-logarithmic velocity profile region is well-defined and ranges from $y^+$ of 30 to 1000 for increasing Reynolds numbers. The data collapse along the Coles equation for the semi-logarithmic region. The wake region is well defined beginning at the point where the velocity profile breaks away from the semi-logarithmic region. Also, there are some data points
in the viscous sublayer, which indicates that we do have flow over a smooth plate. The spectra of the velocity fluctuations has also been obtained and are presented by Ahn (1986).
4.0 INSTRUMENTATION AND EXPERIMENTAL TECHNIQUES

4.1 INSTRUMENTATION

4.1.1 MICROPHONES

The surface pressure fluctuations and cross spectra are measured using small condenser microphones mounted flush to the test section floor. The microphones used are manufactured by Knowles Electronics, Inc. Two different orifice size microphones were used to obtain the data. Model BT-1755 has a relatively large orifice of 1.4 mm in diameter and model BT-1753 has a smaller orifice 0.51 mm in diameter. Both models can be classified as pinhole type microphones for use in these measurements, and are shown in Figure 6. These microphones are used because of their sensitivity, relatively small size and relatively flat response curve over the low frequency range of interest. Figure 7 shows the sensitivity as a function of frequency. From this figure we see that the response is nearly constant at low frequencies from 125 Hz up to approximately 3 kHz. The response peaks near 5 kHz and then decreases with increasing frequency. Also shown are the differences in response at higher frequencies for the two models. The previous experiments done in this wind tunnel by Simpson et al. (1983) used Sennheiser MKH-110 13 mm
diaphragm diameter microphones that were calibrated by the manufacturer. For the present experiments a Sennheiser microphone was used as a base standard microphone for a comparison calibration with the Knowles microphones. Figure 7 shows the manufacturer's response curve for the Sennheiser microphone. The Sennheiser sensitivity is nearly constant between 50 Hz up to 2 kHz.

The Knowles BT-1755 has a rather large orifice for surface pressure measurements. In an effort to reduce inflow and outflow through the pinhole opening, which may be related to the Bull and Thomas effect, these orifices are covered with a small screen as seen in Figure 6. The screen is made by Endevco, Inc. and is used on their models of miniature pressure transducers. Use of these screens did not affect the overall response of the Knowles microphones, but helped provide surface continuity. This can be stated as a result of calibrations performed with and without the screens in place.

4.1.2 MICROPHONE HOUSING UNIT

A unit housing the microphones was designed and used for the measurements. A schematic of the housing unit is shown in Figure 8. The unit houses three microphones, two BT-1755 and one BT-1753. The housing unit containing the microphones was mounted flush with the surface of the flat plate test.
surface and supported from the floor beneath the wind tunnel. The diameter of the unit is 25.4 mm, which is smaller than the hole in the test surface, which is 28.6 mm in diameter. The purpose for this is to prevent vibration from the tunnel contaminating the spectral measurements. The gap left in between is covered with 0.003 cm thick cellophane tape which provides continuity of the surface and yet prevents transmission of vibration to the housing unit. Application of the tape does not contribute to the surface roughness because the thickness of the tape is much smaller than the viscous sublayer.

The housing unit can also be used for cross spectral measurements. One of the BT-1755 microphones is mounted so it can traverse some distance with respect to the other BT-1755 microphone. Using this feature, we can obtain cross spectral data for both the streamwise and spanwise spacings. When obtaining the cross spectral data, the line between centers of the microphones must be aligned parallel or perpendicular to the flow for measurement of the respective cross spectrum. The housing units were used in pairs at each streamwise location, with the same model microphone in each unit used in pairs for measurement of the power spectrum.
4.1.3 CALIBRATION

The Knowles microphones were calibrated using two separate techniques. First a comparison method using a Sennheiser model MKH-110 as a reference microphone, calibrated by the manufacturer, was used for calibration in the frequency range from 4 kHz to 10 kHz and sound pressure levels (SPL) between 60 dB to 119 dB. The response curve for the Sennheiser microphone is shown in Figure 7. A second method using a GenRad model 1986 Omnical Sound Level Calibrator was employed for the frequency range from 125 Hz to 4 kHz. Use of both methods gave a small region of overlap around 4 kHz. Results from both methods gave agreement within 1.5 dB in the overlap region.

The comparison calibration method was performed in a semi-anechoic 1.22 m cubed chamber built by the author. The chamber is constructed of plywood and is lined on the inside with three inch thick acoustic-wave absorbing foam (Sonex "anechoic wedges") that absorbs all energy above 500 Hz. The best results for these calibrations were obtained during hours when the low frequency noise and vibrations from the surroundings were smallest and did not interfere with the calibration. The Sennheiser and Knowles microphones were placed inside of the chamber along with a sound source, a Radio Shack Realistic Super Tweeter, catalog no. 40-1380. A
function generator produced a sinusoidal signal that was fed to an amplifier and then to the speaker.

The low frequency calibration using the GenRad also generated a certain frequency sound at discrete but known SPL. Knowing the SPL and measuring the output voltage one can simply calculate the sensitivity of the microphone in mV/Pa as a function of frequency. Using these calibration method the Knowles microphones showed a response nearly independent of SPL.

The overall calibration of the Knowles microphones showed good agreement within ±1.5 dB of the manufacturer's specifications for all microphones and models used in these experiments. The manufacturer's response curves were used in the data reduction as shown in Figure 7. Simplification of the signal processing was possible since each Knowles microphone of a given model had the same frequency response curve within 1.5 dB.

4.1.4 OTHER INSTRUMENTATION

Additional equipment was used for the power supply, data acquisition, data reduction and plotting. A Hewlett Packard model 6213A power supply was used for the microphones power source. A four channel Data 6000 model 611 and model 681 disk drive by Data Precision was used for acquisition and storage of data. Also a TSI model 1015C correlator was used to add
and subtract the time-varying output signals from the microphones before processing in the data acquisition system. The surface pressure measurements were monitored on a Princeton Applied Research model 4512 FFT Real Time Spectrum Analyzer to insure that data were acquired from well-behaved signals. The data reduction was performed using an IBM PC and IBM 370. The results were plotted using a Hewlett Packard 7475A plotter and Versatec plotter.

4.2 EXPERIMENTAL TECHNIQUES

The measurement of the surface pressure spectra was obtained using the model microphone pairs on the two housing units widely separated in the spanwise direction. The microphone pair of model BT-1755 on each unit was used to measure the cross spectra for both the spanwise and streamwise spatial separations. Through manipulation of the output signals we can obtain a single surface pressure spectrum of only the pressure fluctuations due to the turbulent flow field. This single spectrum does not contain any apparent influence of the acoustic disturbances and flow unsteadiness generated by the blower. This measurement technique was previously used successfully in the experiments performed by Simpson et al. (1983).
4.2.1 DATA ACQUISITION AND SIGNAL PROCESSING

Acoustic waves and flow unsteadiness generated by the tunnel are present and must be accounted for. The acoustic waves and flow unsteadiness are assumed to be the same at a given streamwise location at any instant in time because the tunnel test section acts like a wave-guide. The turbulent spectrum produced by the flow was the same across the test section at a given streamwise location because the mean flow and mean square turbulence structure was two-dimensional in nature across the center of the flow. The acoustic and turbulent signals are uncorrelated since the turbulent pressure fluctuations were generated in a volume local to a measurement position while the inviscid acoustic and unsteady fluctuations were generated far upstream. Equations (7) and (8) show that the turbulence produced is due to the local velocity field. This observation allows us to decompose the surface pressure fluctuations into acoustic and turbulent terms. The two housing units shown in Figure 8 were spaced one-third of a meter apart in the spanwise direction. This distance is greater than 4.5 $\delta$ in the spanwise direction for the thickest boundary layer examined. Therefore, the turbulent pressure signals produced were uncorrelated, yet were statistically the same since the mean flow was 2-D in structure. The decomposition of the time-varying pressure fluctuation signals for a given frequency $n$, is written as
The subscripts a and t designate the acoustic and turbulent pressure fluctuations, and the subscripts 1 and 2 denote the two microphone housing units. Subtracting $p_{2n}$ from $p_{1n}$ we can obtain the mean square value of the turbulent-flow produced pressure fluctuation as a function of frequency, $n$

\[
p_{1tn}^2 = \frac{(p_{1n} - p_{2n})^2}{2}. \quad (17)
\]

This term is the contribution of the turbulent term to the spectrum. The above equation is true because the following conditions exist for the test flows.

\[
p_{1tn}^2 = p_{2tn}^2
\]

(mean 2-D flow)

\[
p_{1an}p_{1tn} = p_{2an}p_{2tn} = p_{1an}p_{2tn} = p_{2an}p_{1tn} = 0
\]

(uncorrelated turbulent and acoustic contributions)

\[
p_{1tn}p_{2tn} = 0
\]

(uncorrelated turbulent contribution)
\[ P_{1an} = P_{2an} \]

(same acoustic signals)

Using the same conditions above, addition of the signals gives the acoustic contribution as a function of frequency for the acoustic spectrum

\[ P_{1an}^2 = \frac{(p_{1n} + p_{2n})^2}{4} - \frac{p_{1tn}^2}{2}. \quad (18) \]

The proper turbulent spectrum is obtained using equation (17) for frequencies below \( c/w \), where \( w \) is the width of the test section. The longitudinal, vertical and spanwise acoustic contributions that are the same at the two microphone units are eliminated using this equation at the same streamwise location. However, anti-symmetric spanwise acoustic contributions near the frequency \( c/w \) and higher harmonics are added to the spectrum. The turbulent contributions for these frequencies are obtained using the following equation

\[ P_{1tn}^2 = \frac{(p_{1n} + p_{2n})^2}{2}. \quad (19) \]
No anti-symmetric spanwise acoustic contributions were observed in these experiments. This led to great simplification in data reduction for the present experiments.

The convective wave speeds and the coherence signals were measured using a like microphone pair, Knowles model BT-1755, on one unit but spaced some small distance apart in either the streamwise or spanwise directions. The wave speed or celerity as a function of \( n \) is given by the following equation

\[
U_{cn} = 2\pi n \Delta x / \phi_n \quad (20)
\]

where

\[
\tan \phi_n = I_n / R_n \quad (21)
\]

and

\[
\gamma^2(\Delta x, n) = \frac{R_n^2 + I_n^2}{n} \quad (22)
\]

Here \( R_n \) is the normalized co-spectrum of the two signals while \( I_n \) is the normalized quadrature; the power spectra of the two signals were used in this normalization. A similar equation can be written for the spanwise direction. Because the acoustic contributions at two different streamwise locations are coherent but time delayed, they can be
accounted for from the measured acoustic spectra. They make negligible contributions to the co-spectrum and quadrature for the experiments reported here.

The signals from each microphone were input into the TSI correlator where addition and subtraction of the time dependent signals were performed. The response of all microphone pairs were nearly identical, therefore the condition that allows us to use equation (17) holds true for all measurements. Data acquisition was performed by the Data 6000 on the output signals of the correlator. The Data 6000 performs a FFT on 0.1 seconds of data for the sum and difference of the two microphone signals. The respective power spectra or cross spectra for 100 successive 0.1 seconds records were averaged to obtain the resultant power spectra or cross spectra, respectively. The raw data were stored on a diskette for additional reduction.

4.2.2 UNCERTAINTY ESTIMATION

The measurement error in the velocity data obtained by the hot-wire anemometer were within ±1 percent uncertainty for the mean velocity and about ±4 percent uncertainty for the rms velocity (Ahn, 1986). The experimental uncertainty for the pressure fluctuations was within ±1.5 dB in spectral level including effects of finite bandwidth and finite record length (Bendat and Piersol, 1971). The uncertainty increases
at the lowest and highest frequencies in the range of interest. The coherence was as much as ±10 percent uncertain for 0.2<φ<20 but at the lowest and highest values of the phase angle, ϕ, the coherence data were more uncertain, by ±0.1. As pointed out on pp. 193-196 of Bendat and Piersol (1971), this uncertainty can be expected for the record lengths used here and for the frequencies with the lowest coherence. Some relatively small uncertainty in the cross spectral data was introduced because of a slight phase difference between microphones which is approximately ±5.5°. Thus, the wave speed data are about ±10 percent uncertain because of the uncertainties in the cross-spectral data.
5.0 EXPERIMENTAL RESULTS

5.1 SURFACE PRESSURE SPECTRA

The results presented are for two different streamwise pressure gradient flows, two with a zero pressure gradient and one with a favorable pressure gradient. For both flows both Knowles BT-1753 and BT-1755 microphones were used. Results vary between the different microphone models. This is reasonable because previous discussion has pointed out the variation in previous results due to different sensing diameters of the transducers. The spectral data presented here are mainly for the results obtained using the smaller orifice microphone, model BT-1753. However, spectral data for the model BT-1755 are presented only for the zero pressure gradient case to illustrate the differences in the results. The Corcos size resolution correction was applied to the data presented here. The spectral data are nondimensionalized to facilitate the task of presentation and comparison. The spectral density are nondimensionalized using the inner wall variables and different combinations of the outer boundary layer variables. Each nondimensional grouping helps display different characteristics of the pressure spectra.
5.1.1 ZERO PRESSURE GRADIENT FLOW RESULTS

First, the results of the zero pressure gradient flow experiments are plotted in Figures 9 through 15. Figure 9 shows the spectra for all momentum Reynolds numbers nondimensionalized on the outer region variables $U_e$, $\delta_1$, and the wall shear stress $\tau_w$. The plot shows that this grouping of variables does not collapse the spectra very well. The region that does collapse well is where the spectra varies like $n^{-1}$ for $1.28 \leq \omega \delta_1/U_e \leq 6$ and only for the momentum Reynolds number greater than 5000. The spectral data also collapse for $0.1 \leq \omega \delta_1/U_e \leq 1.0$ and vary like $n^{-0.7}$. The spectra show that the fluctuations that contain the most energy occur over a broadband of frequencies somewhere between 80 and 5000 Hz. The frequencies below 80 Hz and above 5000 Hz make small contributions to the rms pressure fluctuation.

Figure 10 shows that the grouping of the inner variables collapse the data over a larger range of momentum Reynolds numbers at the higher nondimensional frequencies. In this figure, we see that the spectra collapse for nearly all frequencies shown. Only for the smallest two momentum Reynolds numbers at the middle to higher frequencies do those spectra not collapse well. The region where the spectra vary like $n^{-1}$ can be clearly seen, and exists for the nondimensional frequency $0.1 \leq \omega \nu/U^2_{\tau} \leq 0.5$. This nondimensional plot is perhaps the best since it best shows
the dependency of the wall for nearly all frequencies. Only the large-scale lowest frequency contributions that come from the outer region and are not governed by the wall do not collapse.

Figure 11 and Figure 12 show each individual spectrum in offset plots. In these figures the spectra are nondimensionalized by another set of outer variables, \( U_e \), \( \delta_1 \), and \( q_e \), and were chosen for the offset plots because the spectra do not collapse very well in these coordinates. If the spectra in Figures 10 and 11 were plotted so as to try to collapse the data, we would see that the spectral level at \( \omega \delta_1/U_e = 1.0 \) is at approximately \(-50 \pm 1.5 \) dB. These two figures best show the spectral content for each x-location measured. This figure also demonstrates that there is very little scatter in each individual spectrum. These plots also show a \( n^{-1} \) variation. The spectral content at higher frequencies beyond \( \omega \delta_1/U_e = 7.0 \) varies like \( n^{-5.5} \). One important note is that although the spectra are plotted on three different ordinate scales, the shape of the spectra remains the same and only the dB level changes from ordinate to ordinate.

Figure 13 shows the spectra versus the nondimensional wavenumber \( k \delta_1 \). The measured wave speeds reported below were used to obtain the wavenumber \( k = \omega/U_c \). This figure represents the same coordinates used by Panton and Linebarger (1975). Plotted in this figure are the calculated spectra of Panton.
and Linebarger and measured results for nearly the same Reynolds number. The Reynolds number is defined as $U_\tau^5/\nu$ by Panton and Linebarger. Good agreement can be seen with the $n^{-1}$ or $k^{-1}$ (since $U_c$ is nearly constant) region in the middle wavenumbers. However, agreement becomes poor at the highest and the lowest wavenumbers. Panton and Linebarger's spectra do not fall off as rapidly at the high wavenumbers and at the lower wavenumbers. For the measured spectra, the plot does not extend down below $k \delta_1 < 0.8$ because of the uncertainty in the pressure spectra and convective wave speeds. Panton and Linebarger show more spectral contribution at the highest frequency, and the agreement here is poor.

Figure 14 and Figure 15 show the nondimensional spectra on an offset plot for microphone model BT-1755. There are several similarities and differences in these plots compared with the model BT-1753. The most noticeable difference is a large peak in the spectra. The peak occurs at approximately 5625 Hz for all x-locations, suggesting a microphone dependent effect. The peak first occurs at $\omega \delta_1/U_e = 5$ and then increases to $\omega \delta_1/U_e = 19.5$ with increasing momentum Reynolds number. Another difference is that at frequencies beyond the peak, the spectra vary like $n^{-3}$. The best agreement with data from BT-1753 occurs where $\omega \delta_1/U_e = 1$, with the spectral level at approximately -50±3 dB. A region of $n^{-1}$ exists for most x-locations, and exists between $2.0 \leq \omega \delta_1/U_e \leq 10$. The region is very small at low momentum Reynolds number and increases
in range as momentum Reynolds number increases. This effect was also seen in the results from microphone model BT-1753 in Figure 11 and Figure 12.

Figure 2 and Figure 3 show the results of the rms pressure fluctuations nondimensionalized by the wall shear stress and free-stream dynamic pressure. In the first of these figures we see values of $3.258 \leq p'/\tau_w \leq 3.8$ for transducer BT-1753. The transducer BT-1755 gives values of $2.68 \leq p'/\tau_w \leq 3.17$. The differences in levels is seen by the fact that we have two different orifice size transducers. Further examples of this can be seen in Figure 3 where $p'/q_e$ decreases with increasing $d^+$. This further demonstrates the dependence of rms pressure fluctuation on the sensing diameter. Figure 2 also indicates that $p'/\tau_w$ increases with increasing Reynolds number. Furthermore, Figure 3 also shows that $p'/q_e$ increases with the Reynolds number.

The Corcos correction has been applied to the data and was observed to have a 1 dB effect at $\omega L_1/U_e=0.83$ for the BT-1753 data at $x=1.63$ m. The nondimensional rms pressure fluctuation was also affected. The rms pressure fluctuation increased about 30 percent for the BT-1753 and 50 percent for the BT-1755. This is understandable because the larger the orifice, the larger the correction. However, the Corcos correction did not equalize the rms pressure fluctuations from the two model microphones but only brought them closer in magnitude than before the correction was applied.
5.1.2 FAVORABLE PRESSURE GRADIENT RESULTS

The spectral data presented for the favorable pressure gradient is restricted to microphone model BT-1753 but will include some discussion of the results from model BT-1755. The spectral results are presented in the same nondimensionalized plots as for the zero pressure gradient flow. Figure 16 shows a plot of the measured pressure spectra in the first group of the outer boundary layer variables. This plot indicates that the spectra collapse fairly well using this grouping for $0.1 \leq \omega \delta_1 / U_e < 3.0$. The agreement in the middle range of frequencies is very good and similar in range to the zero pressure gradient results. These variables collapse the lower frequencies better for the favorable pressure gradient flow than the zero pressure gradient flow. For the higher frequencies, we can also see that this group of variables does not collapse the data. These are similar trends between the zero and favorable pressure gradient flows. There are several trends that do not appear in the accelerating flow results; one of them is the existence of a region that varies like $n^{-1}$. The spectral data show a region that varies like $n^{-0.7}$ for $0.7 \leq \omega \delta_1 / U_e \leq 2.0$. Another trend that is different is the frequency at which the spectra begin to rapidly decrease with frequency. This occurs very near $\omega \delta_1 / U_e = 3.0$ while for the zero pressure gradient the fall off occurs at $\omega \delta_1 / U_e = 7.0$ for the larger
Reynolds numbers. Another difference is the variation of the spectra at the highest frequencies. For the favorable pressure gradient flow, there was some variation in the slope of the spectra at the higher frequencies between $n^{-5}$ and $n^{-6}$, and the variation is more negative with increasing Reynolds number. However, for the zero pressure gradient flows the spectral variation remained nearly the same at about $n^{-5.5}$ for all Reynolds numbers.

Figure 17 shows that the spectral data nondimensionalized on the inner boundary layer variables collapse very well. As in the zero pressure gradient case, the inner variables take care of the nondimensionalization for nearly the entire range of frequencies. The lower frequencies do not collapse nearly as well, but this too was observed in the zero pressure gradient case. The only other portion of the data not collapsed is in the middle frequencies for the highest momentum Reynolds number case at $x=4.77$ m.

Figure 18 and Figure 19 show offset plots of the individual spectra nondimensionalized by $U_e$, $\delta_1$, and $q_e$ of outer boundary layer variables. These plots again illustrate the fact that there is little scatter in a given spectrum. Another observation made is the failure of these variables to collapse the data as well as the variables in either Figure 16 or Figure 17. The three different groups of variables used to nondimensionalize the spectra only have the
effect of shifting the spectra along the ordinate and abscissa, and not changing the shape of the spectra. This was done to facilitate the comparison procedure between different Reynolds numbers. The spectral plots for microphone Model BT-1755 were obtained but not presented because model BT-1753 gave nearly the same results for the favorable pressure gradient flow. The spectral peak at 5625 Hz observed in the zero pressure gradient flows for model BT-1755 occurred in the accelerating flow but was not nearly as noticeable and in some instances it appeared to be absent. The reason for this was that the spectral contribution at these frequencies was much smaller than in the zero pressure gradient case. Also this spectral peak became more apparent as the momentum Reynolds number increased.

Figure 5 shows $p'/\tau_w$ versus $Re_{\delta_1}$. We observe from this plot that the nondimensionalized rms pressure fluctuation has a value between 2.6 and 2.9 for model BT-1753 and 2.3 and 2.9 for model BT-1755. These levels are very nearly the same for both microphone models. There is a slight increase of $p'/\tau_w$ with an increase of $Re_{\delta_1}$ for BT-1753, but BT-1755 shows a slight decrease with increasing $Re_{\delta_1}$. The plot of $p'/q_e$ versus $d^+$ is shown in Figure 3. Shown here are similar trends discussed for the zero pressure gradient flows. At smaller $d^+$, the value of $p'/q_e$ is the largest. However, for the model BT-1755, the value of $p'/q_e$ is rather high but is in better agreement with previous research than the zero
pressure gradient flow data from model BT-1755. There seems to be some obvious differences in the zero and favorable pressure gradient flows between the different model microphones.

Corcos' correction was also applied to the spectral data for the favorable pressure gradient flow. The correction was observed to have a 1 dB effect at $\omega \delta_1/U_1=0.88$ for the BT-1753 data at $x=1.63$ m. The most noticeable observation was the fact that Corcos' correction brought the rms pressure fluctuation from both model microphones very close in agreement, but the uncorrected data were originally very close in magnitude. This smaller difference in rms pressure fluctuation occurs only for the accelerating flow. This indicates that differences in transducer size is very small in this particular experiment. Application of the Corcos' correction to the two different model microphones did not greatly increase the higher frequency components, thus the rms pressure fluctuation did not increase as much as in the zero pressure gradient case.

5.2 SQUARE ROOT OF THE COHERENCE AND CONVECTIVE WAVE SPEED RESULTS

A pair of model BT-1755 microphones was used to obtain all cross spectral data. The co-spectrum and quadrature were obtained and then the wave speeds and coherence were
extracted from the experimental data. The cross-spectral data were obtained at 10 different microphone spacings in either the lateral, Δz, or longitudinal, Δx, direction. The closest spacing in either direction was 2.413 mm and the largest spacing was 9.172 mm. An attempt was made to obtain data over an even distribution of microphone spacings. The 10 different spacings gave an adequate number of data points to determine both the coherence and the convective wave speeds. Tables 5 and 6 give the values of the microphone spacings for both the zero and favorable pressure gradient flows, respectively.

Figure 20 through Figure 24 show the square root of the coherence obtained in the longitudinal direction in the zero pressure gradient flow. As mentioned in Chapter 2, the square root of the coherence decays approximately like $e^{-K_1}$, where $K_1$ is a decay constant. The decay constant, $K_1$, is 0.2 for the lower Reynolds numbers and increases to 0.3 at the higher Reynolds numbers. Only data between $0.75<\Delta x/\delta_1<2.7$ were obtained at low Reynolds numbers; at high Reynolds numbers $0.20<\Delta x/\delta_1<0.74$. The lateral cross spectra are plotted in Figures 25 through 29. $K_3$ is approximately 0.7. Values of $\Delta z/\delta_1$ were close to those for $\Delta x/\delta_1$. The parameters that collapse the data for all microphone spacings are $\omega\Delta x/U_C$ and $\omega\Delta z/U_C$. The exponential model does not fit the data as well as these parameters collapse the data,
although at higher $\phi$ values where the coherence is low it is also uncertain by ±0.1.

The square root of the coherence for the accelerating flow are shown in Figures 30 through 37. The exponential decay constant, $K_1$ is between 0.1 and 0.2 for the longitudinal direction and decreases with increasing Reynolds number. Values of $\Delta x/\delta_1$ and $\Delta z/\delta_1$ are between 0.75 and 3.35 and increase slightly with Reynolds number. Comparing to the zero pressure gradient flow, the streamwise coherence does not decay as fast. This means that the streamwise extent of the pressure fluctuations remain more coherent in an accelerating flow for larger downstream distances. The decay constant, $K_3$ is between 0.35 and 0.6 for the lateral coherence. $K_3$ increases with increasing Reynolds number, which means that the spanwise extent of the pressure fluctuations become less coherent in the spanwise direction as Reynolds number increases. $K_3$ is not as large as the decay in the zero pressure gradient flow. This further indicates that the non-dimensional spanwise extent of the pressure fluctuation producing flow structures is greater in the accelerating flow. This means that the large-scale structures are slower to change character or shape in the favorable pressure gradient flow than the zero pressure gradient flow. All cross spectral data showed that the decay in the cross spectra was best defined by the smaller microphone spacings between 2.413 mm and 5.11 mm.
The decay of the square root of the coherence was not as well defined for the larger spacings. The coherence does not go to zero for the largest phase angles mainly because of the experimental uncertainties. This has been observed by all earlier researchers. Both the lateral and longitudinal direction results show this trend but it is much more apparent in the favorable pressure gradient case.

The convective wave speeds are shown in Figures 38 and 39. The wave speed is nondimensionalized by the free-stream velocity and plotted versus $\omega \delta_1 / U_e$. In these figures the curve shown is a mean curve for all the results of the present experiment at each x-location for all 10 microphone spacings. The results were not a function of microphone spacing. In both types of flow the wave speed increases with increasing $\omega \delta_1 / U_e$ until some maximum is reached and then the wave speed remains nearly constant. In the zero pressure gradient flow at the higher Reynolds numbers, the wave speed reaches a maximum, then decreases slightly where the wave speed then reaches a nearly constant value. This trend also appears to be true for the lower Reynolds numbers in the accelerating flow.

The ratio of $U_c / U_e$ at high values of $\omega \delta_1 / U_e$ remains nearly the same for all momentum Reynolds numbers in the zero pressure gradient case. However, for the favorable pressure gradient flow the ratio of $U_c / U_e$ at high values of $\omega \delta_1 / U_e$ decreases with increasing Reynolds number.
In the zero pressure gradient flow, the wave speed at higher frequencies remains constant between 56 and 50 percent of the free stream velocity for $\omega \delta_1/U_e \geq 0.5$ with $U_c/U_e$ decreasing with increasing Re. For the favorable pressure gradient case the ratio of $U_c/U_e$ remains nearly constant when $\omega \delta_1/U_e \geq 5$ and ranges in value between 64 and 53 percent of the free stream velocity. The level where $U_c/U_e$ is constant, decreases with increasing Re. The wave speed data presented for each x-location is a mean curve of the data from all ten $\Delta x$ spacings. The scatter in the data, which is not shown in the figure, is about ±10 percent for each streamwise location in the mid-frequency range. The most scatter occurs for the smallest and largest values of $\omega \delta_1/U_e$. Therefore, the data were not presented at these values of $\omega \delta_1/U_e$. 
6.0 DISCUSSION

In the previous chapter, only a discussion of the present results was given. In the present chapter, the discussion centers on comparisons with previous studies.

6.1 SURFACE PRESSURE SPECTRA COMPARISONS

The Introduction and Tables 1 and 2 give a brief overview of the results from previous experiments. Tables 3 and 4 are results from the present studies. First, in comparing the results of the zero pressure gradient flow with others, we can examine the approximate level of the spectral data. The level of the spectra at \( \omega \delta_1/U_e = 1.0 \) gives the best point of comparison because the low frequency noise and high frequency resolution problems are small. For all studies including the present, the spectral level is very near \(-50 \pm 3 \) dB, where \( \text{dB} = 10 \log_{10} \left| \phi(\omega)U_e/q_e^2 \delta_1 \right| \). As discussed in Chapter 5, this grouping of variables does not collapse the data well, but is used because most of the previous results were nondimensionalized by this group of variables. The agreement is good for a wide range of momentum Reynolds numbers. The shape factor, \( H \) is approximately 1.3 and \( 0.031 \leq U_t/U_e \leq 0.04 \), indicating similar boundary layer characteristics for nearly all other previous research.
Many of the differences among different studies can be attributed to the microphone resolution effect as shown in Figure 3. The present study has the best resolution and the smallest value of $d^+$. The present data are also plotted here and show excellent agreement with Blake (1970), Dinkelacker et al. (1977) and Emmerling (1972) all of whom used transducers with values of $d^+$ close in magnitude to the present study. Bull and Thomas (1976) show values of $p'/q_e$ some 30 percent lower in magnitude than the consensus of other experiments. Bull and Thomas claim their data are more correct and that Blake's results were too high. As the present study, Blake used pinhole microphones. However, it is difficult to say the pinhole effect has caused values of $p'/q_e$ to be too high because the data from Dinkelacker and Langeheineken (1982), Emmerling (1973) and Schewe (1982), who used flush mounted surface transducers, are in agreement with Blake (1970) and the present data.

The Bull and Thomas correction was applied to some of the present data and was observed to overcorrect the data. This can be stated because for the present data values of $d^+$ are smaller than for Bull and Thomas and when the correction was applied the value of $p'/q_e$ was nearly the same when compared to the values given by Bull and Thomas. However, we know that the ratio of $p'/q_e$ increases with decreasing $d^+$. Thus, it is the opinion of the present researchers that the Bull and Thomas effect may exist, but is not as large as
claimed by Bull and Thomas. For the present data, there are
distinct differences between the values of $p'/q_e$ for the two
different pinhole sizes. The lower $p'/q_e$ at the larger $d^+$
for model BT-1755 is seen in Figure 3. The effect of transducer size is clearly shown by the past and present
data.

Figure 10 shows the mean spectra line from Bull and
Thomas (1976) for the zero pressure gradient flow. Spectral
shape and distribution are in fair agreement. The part that
least agrees is in the region $0.3 \leq \omega \nu/U_1^2 \leq 0.7$. This region
is where Bull and Thomas showed the pinhole effect to be most
significant. The difference in spectral level is between 2
to 3 dB in the mid to high frequency range.

For the favorable pressure gradient flow, the consensus
level of the spectra at $\omega \delta_1/U_1^2 = 1$ is at approximately $49 \pm 2$ dB,
where $dB = 10 \log_{10} |\phi(\omega)U_1/q_e^2 \delta_1|$. The spectral level for the
favorable pressure gradient appears to be about 1 dB higher
than the zero pressure gradient. The favorable pressure
gradient flows have been performed over a large range of $Re_\theta$
and $d^+$. $U_1/U_e$ ranges from 0.04 to 0.05 for all the
accelerating flows. The shape factor, $H$ is approximately 1.3
for most of the previous studies as well as the present
study.

The data of Burton (1973) and the present data show that
there is a small affect on $p'/q_e$ due to the streamwise
pressure gradient at small values of \( d^+ \). The trend seems to indicate that \( p'/q_e \) is relatively constant for small \( d^+ \).

Figure 4 shows several mean spectra for Schloemer (1967), Blake (1970), Burton (1973), and the present study. Spectra from both zero and favorable pressure gradients are shown in this figure. These curves represent mean curves of the results for a particular study. The levels of the spectra are nearly the same at \( \omega \delta_1/U_e = 1.0 \) for different streamwise pressure gradients. Shown best on this plot is the difference in high frequency spectral content. The zero pressure gradient flow spectra show much more spectral content at the higher frequencies than the favorable pressure gradient flow spectra. However, when the spectra are nondimensionalized on \( v \) and \( U_1 \) we observe a collapse of the data independent of streamwise pressure gradient, indicating that this grouping of inner variables will collapse the spectral data best. At smaller Reynolds numbers for the favorable pressure gradient flow there is less energy at the higher frequencies when compared to the zero pressure gradient flow.

Another important trend in the data deals with the variation of the spectrum when values of \( \omega \delta_1/U_e \) are between 1 and 28. Bradshaw (1967) indicates that spectra should vary like \( n^{-1} \) in an overlap region and Panton and Linebarger (1973) use this to predict their spectral data as a function of wavenumber, \( k \). The overlap region exists in the spectra
where the inner wall variables and outer variables both scale flow phenomena and each can be used to nondimensionalize and collapse the surface pressure spectral data. The present results for the zero pressure gradient flow show the existence of this region between $1 \leq \omega \delta_1 / U_e < 6$ for $Re_\theta > 5000$. For $Re_\theta < 5000$ the zero and favorable pressure gradient flow show a variation more like $n^{-0.7}$. The predicted overlap region is quite large and is shown to exist for over a decade by Panton and Linebarger (1973). However, the data presented here along with others show a much shorter region. Spectra presented by Schloemer (1967) and Burton (1973) do not show the existence of the overlap region for either the zero or favorable pressure gradient flows, but their $Re_\theta < 5000$. Other researchers show a $n^{-1}$ region ranging from $1 \leq \omega \delta_1 / U_e \leq 20$. Bull (1967) shows the largest region. Bull (1967), Schewe (1982) and Dinkelacker et al. (1977) also show the existence of the $n^{-1}$ in duct flows. The frequency range of the overlap region increases in with $Re_\theta$. Further comparison can be seen in Figure 13, in the coordinates $10 \log_{10} |\phi(k) / \omega_1^2 \delta_1|$ versus $k \delta_1$. Panton and Linebarger's calculation is plotted along with the present results and we can see the overlap region. The present data only have the $k^{-1}$ region between $4.0 \leq k \delta_1 \leq 10$ but Panton and Linebarger show the region between $2.5 \leq k \delta_1 \leq 30$. The spectra are not plotted below $k \delta_1 < 0.8$ because of large uncertainties in the frequency content and convective wave speeds in the present data. Also, the higher wave numbers
show a faster drop off than do Panton and Linebarger. Panton and Linebarger also observed this when comparing their calculations to the experimental results of Willmarth and Roos (1965) and suggested that the transducer resolution is the reason for this faster drop off.

At the higher frequencies the present results show that the spectrum varies like $n^{-5.5}$. This fall off in spectral content is due to either a real effect of the flow or by the resolution of the transducer. Previous Corcos corrected results with various diameter transducers show a variation like $n^{-3}$ to $n^{-4}$, which is not as steep as for the present results but indicates that the surface pressure spectra may drop off fast. This means that the drop off in the spectral content may be a real occurrence and not due to resolution of the transducers. The present favorable pressure gradient flow also shows a rapid drop off of $n^{-5}$ to $n^{-6}$ in spectral content for the higher frequencies while previous studies show a spectral variation like $n^{-3.3}$ to $n^{-4}$. However, there is some indication that the spectral drop off is real and not just due to the resolution problems.

Figure 2 and Figure 5 show $p'/\tau_w$ versus $Re_{\delta_1}$. These figures contain the results of several studies, including the present one. Both figures show the slight Reynolds number dependence on $p'/\tau_w$. All studies shown in Figure 2 indicate that there is a slight increase of $p'/\tau_w$ with an increase in $Re_{\delta_1}$. Figure 5 shows the dependence of $p'/\tau_w$ on
transducer size. For the zero pressure gradient flow, Blake (1970) and the present study agree best. However, both use pinhole type microphones which Bull and Thomas (1976) claim make the results too large. However, in support of the results of this present study, Lim (1971) states that Hodgson had indicated that $p'/\tau_w>4$ is more correct for a zero pressure gradient flow. The present pinhole data is closest in agreement to Hodgson's value.

$p'/\tau_w$ in the favorable pressure gradient is lower in value than the zero pressure gradient. The data plotted in Figure 5 show fairly good agreement between Schewe (1982), Burton (1973) and the present data. In this figure the data by Schewe (1982) show the affect of $d^+$, but indicate that the smallest value of $d^+$ gives the highest and most reasonable value of $p'/\tau_w$. It is difficult to say what the value of $p'/\tau_w$ should be for the favorable pressure gradient case. The slight Re dependence for the zero pressure gradient flow shows that if you extend a line following the trend down to the lower values of Re, $p'/\tau_w$ should be somewhere between 3 and 3.2. This is reasonable since the law-of-the-wall velocity and turbulence structure describes both zero and favorable pressure gradient flows at low Reynolds numbers. Using this as a reference for the level in the favorable pressure gradient flow, the data presented agrees well for similar $d^+$ for this level of $p'/\tau_w$. The good agreement for the different microphone models used in the present
experiments in the favorable pressure gradient is observed in Figure 5 because the high frequency content in the surface pressure spectra is not as large in the favorable pressure gradient flow as compared to the zero pressure gradient flow. Therefore, the resolution problems are not nearly as large, at least not for the lower Re in the favorable pressure gradient flow. The larger values of Re show that there is less agreement between the two different size microphones, indicating that the resolution has an affect at these Reynolds numbers. The resolution issue seems to be the best explanation for the agreement between results for different pinhole sizes in the favorable pressure gradient flow and the reason for the large differences in the zero pressure gradient flows.

6.2 SQUARE ROOT OF THE COHERENCE AND CONVECTIVE WAVE SPEED COMPARISONS

Shown previously in Chapter 2 was the exponential decay model equation (13), used by Corcos (1963) and Brooks and Hodgson (1981) to fit their coherence data. Observing the data of Schloemer (1967), Blake (1970), Burton (1973), Bull (1967) and others, we see that their coherence data also decay almost exponentially. In general, the present results seem to follow an exponential decay for $\phi<5$. Tables 1 through 4 show the values of the decay constants $K_1$ and $K_3$.
for past experiments for $\Delta x/\delta > 3$ and present experiments. In general, the exponential decay model does not fit the data very well.

Schloemer (1967) first examined the differences in the coherence due to the streamwise pressure gradient. Using the zero pressure gradient case as a basis for comparison, Schloemer indicated that for favorable pressure gradients, the streamwise decay of coherence is slower than for the zero pressure gradient case. However, in an adverse pressure gradient flow, the decay is more rapid. Schloemer also suggested that there is little difference in the lateral decay due to the streamwise pressure gradient. Examining the results of the exponential decay constant, one observes that in most cases this statement is true. In the zero pressure gradient flow the present results gave $0.2 \leq K_1 \leq 0.3$ and $K_3 = 0.715$, which shows good agreement with earlier results for $\Delta x/\delta < 2$. Here, as in earlier experiments, the values of $K_1$ increase with decreasing $\Delta x/\delta$ and increasing Reynolds number while $K_3$ remains constant with decreasing $\Delta x/\delta$ and increasing Reynolds number. In the favorable pressure gradient flow the present results gave $0.1 \leq K_1 \leq 0.2$ and $0.35 \leq K_3 \leq 0.6$, which shows good agreement with previous research for small $\Delta x/\delta$ values. Here values of $K_1$ increase slightly with decreasing $\Delta x/\delta$ and increasing Reynolds number while $K_3$ increases with decreasing $\Delta z/\delta$ and increasing Reynolds number.
It is difficult to determine trends in the previous data for both streamwise pressure gradients because there was little work done over the range of variables. Considering Schloemer's data and his statement above, we can observe that there is little effect due to the pressure gradient. The values of $K_1$ are nearly the same for both streamwise pressure gradient flows. However, the values of $K_3$ seem to decrease for the favorable pressure gradient flows. This means that the spanwise extent of the pressure fluctuations in the favorable gradients remains larger as the pressure fluctuations move downstream when compared to the zero pressure gradient flow. Brooks and Hodgson (1981) had a weak adverse pressure gradient and their value of $K_3$ is in agreement with the favorable pressure gradient flows. The value of $K_1$ for Brooks and Hodgson show good agreement with both the present zero and favorable pressure gradient flows for some values of $\Delta x/\delta_1$ and $\Delta z/\delta_1$. This seems to indicate that whether the flow is accelerating or decelerating, the longitudinal cross spectra and coherence decay about the same as the zero pressure gradient flow. Also, the lateral cross spectra and coherence do not seem to decay as fast for an accelerating or decelerating flow when compared to the zero pressure gradient flow.

The statement made by Schloemer and discussed above is supported by a paper by White (1964). The paper states that theoretically the longitudinal cross spectral level is
higher, or the decay in the favorable pressure gradient flow is slower than for an adverse or zero gradient pressure gradient flow. White's theory shows little difference in the lateral cross spectral level or decay as a result of the streamwise pressure gradient.

The exponential decay model fits the cross spectral data at the lower values of the phase angle, $\phi$, as discussed earlier in this section. The exponential decay model goes to zero quite fast for the values of $\phi>5$, but the square root of the coherence for the present data does not decay to zero for $\phi>5$ for either the spanwise or streamwise direction. The coherence only decays to some level where it then remains nearly constant for increasing phase angle, $\phi$. Coherence results for all previous researchers show that when $\phi \geq 5$ the coherence also does not go exactly to zero. This occurs at the larger values of the phase angle because the large scale structures in the boundary layer add to the apparent coherence raising the level of the cross spectra, and do not show a decay in the coherence to zero. This occurs in both the streamwise and spanwise directions.

The convective wave speeds are plotted in Figures 38 and 39 for the zero and favorable gradients flows respectively. $U_c/U_e$ increases with increasing $\omega \delta_1/U_e$ up to $\omega \delta_1/U_e \geq 0.5$ where $U_c/U_e$ then becomes constant in the zero pressure gradient flow; in the favorable pressure gradient flow $U_c/U_e$ is constant near 0.6 for $\omega \delta_1/U_e \geq 5.0$. Blake (1970) and Burton
(1973) show similar results. However, for Blake's zero pressure gradient flow and Burton's accelerating flow, $U_c/U_e$ is constant when $\omega \delta_1/U_e \geq 2.0$. These three studies also show that $U_c/U_e$ reaches a maximum or peak near $\omega \delta_1/U_e = 0.5$, then decreases slightly before reaching a constant value of $U_c/U_e$. Schloemer (1967) and Bull (1967) show different results of $U_c/U_e$ for small values of $\omega \delta_1/U_e$. Their data indicate that $U_c/U_e$ decreases with increasing $\omega \delta_1/U_e$ and then becomes constant for $\omega \delta_1/U_e > 2$; values of $U_c/U_e = 0.6$ and 0.8 were observed for zero and favorable pressure gradients. Schloemer (1967) also shows a dependence on microphone spacing while Burton (1973) does not. Brooks and Hodgson also show $U_c/U_e$ as a function of spacing. The wave speeds of individual motion are not a function of sensor spacing. Because as the pressure fluctuation producing motions move downstream the slower near-wall small-scale effects die out rather quickly but the faster large-scale motions remain more coherent, whereby indicating a false impression that the convection velocity for a given frequency increases with microphone spatial separation. This effect is also a possible reason why the coherence does not completely decay to zero for large values of $\phi$.

Although there is some large degree of uncertainty in the present results of the convective wave speeds, the agreement with previous research is good for both streamwise pressure gradient flows.
7.0 CONCLUSIONS

Here it has been demonstrated that a new experimental technique using two microphones spaced far apart in the spanwise direction was successful in obtaining reasonable and consistent surface pressure fluctuation results for zero and favorable pressure gradient flows.

This investigation provides extensive documentation of the spectral trends and levels for both streamwise pressure gradients. Good agreement was obtained with previous results. \( p'/\tau_w \) and \( p'/q_e \) are functions of \( \text{Re}_e \) and \( d^+ \).

For the zero pressure gradient flow the spectra at low \( \text{Re}_e \) collapse on the plot of \( 10 \log_{10} |\phi(\omega)U_e/\tau_w^2\delta_1| \) versus \( \omega\delta_1/U_e \) for \( 0.1 \leq \omega\delta_1/U_e \leq 1.0 \) with \( n^{-0.7} \). For \( \text{Re}_e > 5000 \) the spectra collapse on the plot of \( 10 \log_{10} |\phi(\omega)U_e/\tau_w^2\delta_1| \) versus \( \omega\delta_1/U_e \) for \( 1.28 \leq \omega\delta_1/U_e \leq 6.0 \) with \( n^{-1.0} \). Also for the zero pressure gradient flow the spectra collapse on the plot of \( 10 \log_{10} |\phi(\omega)/\rho^2U_t^2\nu| \) versus \( \omega\nu/U_t^2 \) for \( \omega\nu/U_t^2 \geq 0.1 \). For \( 0.05 \leq \omega\nu/U_t^2 < 0.1 \), the overlap region varies like \( n^{-1.0} \) and at the higher frequencies the spectra varies like \( n^{-5.5} \).

For the favorable pressure gradient flow, which occurs at low \( \text{Re}_e \), the spectra collapse on the plot of \( 10 \log_{10} |\phi(\omega)U_e/\tau_w^2\delta_1| \) versus \( \omega\delta_1/U_e \) for \( 0.1 \leq \omega\delta_1/U_e \leq 3.0 \) and this region varies like \( n^{-0.7} \). Also for the favorable pressure gradient flow the spectra collapse on the plot of \( 10 \log_{10} |\phi(\omega)/\rho^2U_t^2\nu| \) versus \( \omega\nu/U_t^2 \).
10\log_{10}|\phi(\omega)/\rho^2U^2_\nu| \text{ versus } \omega v/Un^2 \text{ for } \omega v/U^2 n^2 \geq 0.3. \text{ For } 0.01 \leq \omega v/U^2 n^2 < 0.3, \text{ the region varies like } n^{-0.7} \text{ and for the higher frequencies the spectra varies like } n^{-5} \text{ to } n^{-6}.

The spectral data agree well with the calculation method of Panton and Linebarger for the zero pressure gradient flows. Their calculation method does not include accelerating flows, therefore, no comparisons were made. The Bull and Thomas pinhole effect seems to overcorrect when applied to the present data. The Corcos correction seems to correct most of the resolution problems at the higher frequencies where the wavelengths of pressure fluctuation are on the order of the sensing diameter. The best results for the surface pressure fluctuation spectra were obtained using the smaller orifice diameter microphone model BT-1753.

The square root of the coherence demonstrates an approximate exponential decay for small values of the phase angle, \(\phi\) as seen in previous studies. Good agreement with earlier values of the exponential decay constants \(K_1\) and \(K_3\) were observed for \(\Delta x/\delta_1 < 3\). The decay of the coherence is defined best by the smaller spatial separations for both streamwise pressure gradient flows. The longitudinal and lateral coherence do not decay to zero because of experimental uncertainties and because the large-scale low frequency structures may make the coherence remain at some finite level for large values of \(\phi\). The longitudinal coherence decays at about the same rate for both streamwise
pressure gradients. The lateral coherence decays faster in a zero pressure gradient flow than a favorable pressure gradient flow because for the accelerating flow the large scale structures tend to scale on the upstream boundary layer thickness, so the spanwise extent of the correlation remains larger in terms of $\omega \Delta z / U_c$. The longitudinal and lateral coherence tend to collapse for all 10 microphone spacings for all cases in the present study.

Good agreement with previous research for the convective wave speeds was shown. For the zero pressure gradient flow, the ratio of $U_c / U_e$ is near 0.5 for large values of $\omega \delta_1 / U_e$ for values of $Re_\theta > 5000$. For the favorable pressure gradient flow, the ratio of $U_c / U_e$ is near 0.6 for large values of $\omega \delta_1 / U_e$ for values of $Re_\theta > 2440$. 

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Figure 1. Side View of the Wind Tunnel Test Section: Solid line is the contour for dP/dx=0 flow and dashed line is the contour for dP/dx<0 flow. Major divisions shown by rule are 10 inches.
Figure 2. Zero Pressure Gradient Flow Results for $p'/\tau_w$ versus $Re_{\delta}$: Solid line, Panton and Linebarger (1974) from equations (11) and (12); ○ Schloemer (1967); ● Present Data model BT-1753; ■ Present Data model BT-1755; Dashed line, Blake (1970); Line with symbols, Bull and Thomas (1976); △ Lim (1971); Shaded region, Bull (1967) and Willmarth (1958).
Figure 3. \( p'/q \) versus \( dU_t/\nu \) for the Zero and Favorable Pressure Gradient Flows: □ Willmarth and Roos (1965) \( dP/dx=0 \); ▽ Bull (1967) \( dP/dx<0 \); × Emmerling (1972) \( dP/dx<0 \); ◊ Bull and Thomas (1976) \( dP/dx=0 \); △ Langeheineken and Dinklelacker (1982) \( dP/dx<0 \); ● Schewe (1982) \( dP/dx<0 \); ▼ Schloemer (1967) \( dP/dx=0 \); ○ Blake (1970) \( dP/dx=0 \); ▲ Lim (1971) \( dP/dx=0 \); ◆ Burton (1973) \( dP/dx<0 \); ◄ Present Data model BT-1753 \( dP/dx=0 \); ■ Present Data model BT-1755 \( dP/dx=0 \); ▼ Present Data model BT-1753 \( dP/dx<0 \); ▲ Present Data model BT-1755 \( dP/dx<0 \).
Figure 4. Mean Curves of the Nondimensional Pressure Spectra for Several Researchers in Both Zero and Favorable Pressure Gradients: Solid line, Burton (1973) dP/dx=0; Dashed line, Blake (1970) dP/dx=0; ○ Schloemer (1967) dP/dx=0; △ Present Data, model BT-1753 dP/dx=0; ×× Schloemer (1967) dP/dx<0; ■ Burton (1973) dP/dx<0; ▽ Present Data, model BT-1753 dP/dx<0.
Figure 5. Favorable Pressure Gradient Flow Results for $p'/t_w$ versus $Re_{\delta_1}$: ● Schloemer (1967); □ Burton (1973); ▲ Schewe (1982), for five different values of $d^+$; ○ Present Data model BT-1753; ■ Present Data model BT-1755.
Figure 6. Schematic of the Endevco Microphone Screen and Knowles Electronics Microphones.
Figure 7. Microphone Sensitivity (V/Pa) versus Frequency (n): The solid line is for microphone model BT-1753, the dashed line is for microphone model BT-1755 and the solid line with symbols is for the Sennheiser model microphone supplied by the manufacturer.
Figure 8. Schematic of the Microphone Housing Unit: Note that two BT-1755 and one BT-1753 microphones are in each unit.
Figure 9. Nondimensional Pressure Spectra Normalized on $\delta_1$, $U_e$ and the wall shear stress, $\tau_w$, the Outer Variables for the Zero Pressure Gradient Flow: Results for BT-1753 at the following $x$-locations, • 1.63 m; ◇ 1.88 m; ▲ 2.22 m; ▲ 2.54 m; ▲ 2.86 m; ▼ 3.52 m; △ 4.14 m; ◆ 5.48 m; ◊ 6.51 m; ◊ 6.51 m (q=2.4"H$_2$O).
Figure 10. Nondimensional Pressure Spectra Normalized on $v$ and $U_\tau$, the Inner Variables for the Zero Pressure Gradient Flow: Results for BT-1753 at the following $x$-locations, $\bullet$ 1.63 m; $\bullet$ 1.88 m; □ 2.22 m; ▲ 2.54 m; ⬇ 2.86 m; ▼ 3.52 m; ◂ 4.14 m; ○ 5.48 m; ◆ 6.51 m; □ 6.51 m ($q=2.4''H_2O$); Dashed line, Bull and Thomas (1976).
Figure 11. Nondimensional Pressure Spectra Normalized on $\delta_1$, $U_e$ and $q_e$, the Outer Variables for the Zero Pressure Gradient Flow: Results for BT-1753 at the following x-locations, • 1.63 m; ● 1.88 m; ■ 2.22 m; ▲ 2.54 m; □ 2.86 m. The solid line is where the spectra varies like $n^{-1}$, the dashed line is a $n^{-5.5}$ variation, and the solid line with symbols is a $n^{-0.7}$ variation. Note the offset of each curve from top to bottom of 10 dB.
Figure 12. Nondimensional Pressure Spectra Normalized on $\delta_1$, $U_e$ and $q_e$, the Outer Variables for the Zero Pressure Gradient Flow: Results for BT-1753 at the following x-locations, ▼ 3.52 m; ▲ 4.14 m; ○ 5.48 m; ◊ 6.51 m; □ 6.51 m (q=2.4"H$_2$O). The solid line is where the spectra varies like $n^{-1}$, and the dashed line is a $n^{-5.5}$ variation. Note the offset of each curve from top to bottom of 10 dB.
Figure 13. Nondimensional Pressure Spectra as a function of Wavenumber for the Zero Pressure Gradient Flow: Results for BT-1753, $\bullet$ 1.63 m, $U_t \delta/\nu=1350$ and for Panton and Linebarger $U_t \delta/\nu=1000$;
$\diamond$ 6.51 m, $U_t \delta/\nu=3876$ and for Panton and Linebarger $U_t \delta/\nu=4000$. 
Figure 14. Nondimensional Pressure Spectra Normalized on $\delta_1', U_e$ and $q_e'$, the Outer Variables for the Zero Pressure Gradient Flow: Results for BT-1755 at the following x-locations, $\bullet$ 1.63 m; $\bullet$ 1.88 m; $\blacksquare$ 2.22 m; $\blacktriangle$ 2.54 m; $\blacklozenge$ 2.86 m. The solid line is where the spectra varies like $n^{-1}$, the dashed line is a $n^{-5.5}$ variation, and the solid line with symbols is a $n^{-3}$ variation. Note the offset of each curve from top to bottom of 10 dB.
Figure 15. Nondimensional Pressure Spectra Normalized on $\delta_1$, $U_e$ and $q_e$, the Outer Variables for the Zero Pressure Gradient Flow: Results for BT-1755 at the following x-locations, ▼ 3.52 m; ▼ 4.14 m; O 5.48 m; ◇ 6.51 m; □ 6.51 m ($q=2.4''H_2O$). The solid line is where the spectra varies like $n^{-1}$ and the dashed line is a $n^{-3}$ variation. Note the offset of each curve from top to bottom of 10 dB.
Figure 16. Nondimensional Pressure Spectra Normalized on \( \delta_1, U_e \) and the wall shear stress, \( \tau_w \), the Outer Variables for the Favorable Pressure Gradient Flow: Results for BT-1753 at the following x-locations, ● 1.63 m; ○ 1.88 m; ■ 2.22 m; ▲ 2.54 m; ● 2.86 m; △ 3.12 m; ▼ 3.52 m; ● 4.14 m; ▽ 4.77 m.
Figure 17. Nondimensional Pressure Spectra Normalized on $v$ and $U_T$, the Inner Variables for the Favorable Pressure Gradient Flow: Results for BT-1753 at the following x-locations, $\bullet$ 1.63 m; $\bullet$ 1.88 m; $\blacksquare$ 2.22 m; $\blacktriangle$ 2.54 m; $\blacktriangledown$ 2.86 m; $\blacktriangleleft$ 3.12 m; $\blacksquare$ 3.52 m; $\bigcirc$ 4.14 m; $\blacktriangle$ 4.77 m.
Figure 18. Nondimensional Pressure Spectra Normalized on $\delta_1$, $U_e$ and $q_e$, the Outer Variables for the Favorable Pressure Gradient Flow: Results for BT-1753 at the following x-locations, $\bullet$ 1.63 m; $\Diamond$ 1.88 m; $\blacksquare$ 2.22 m; $\triangle$ 2.54 m; $\Box$ 2.86 m. The solid line is where the spectra varies like $n^{-0.7}$, and the dashed line is a $n^{-5}$ variation. Note the offset on each curve from top to bottom of 10 dB.
Figure 19. Nondimensional Pressure Spectra Normalized on $\delta_1$, $U_e$ and $q_e$, the Outer Variables for the Favorable Pressure Gradient Flow: Results for BT-1753 at the following x-locations, $\Delta$ 3.12 m; $\nabla$ 3.52 m; $\triangledown$ 4.14 m; $\triangledown$ 4.77 m. The solid line is where the spectra varies like $n^{-0.7}$, and the dashed line is a $n^{-6}$ variation. Note the offset on each curve from top to bottom of 10 dB.
Figure 20. Longitudinal y for the Zero Pressure Gradient Flow, x=1.63 m: Solid line is the exponential decay with $K_1=0.2$. 
Figure 21. Longitudinal $y$ for the Zero Pressure Gradient Flow, $x=3.52$ m: Solid line is the exponential decay with $K_1=0.2$. 
Figure 22. Longitudinal $\gamma$ for the Zero Pressure Gradient Flow, $x=4.14$ m: Solid line is the exponential decay with $K_1=0.3$. 
Figure 23. Longitudinal $y$ for the Zero Pressure Gradient Flow, $x=6.51$ m: Solid line is the exponential with $K_1=0.3$. 
Figure 24. Longitudinal $\xi$ for the Zero Pressure Gradient Flow, $x=6.51$ m ($q=2.4''H_2O$): Solid line is the exponential with $K_1=0.3$. 
Figure 25. Lateral $\gamma$ for the Zero Pressure Gradient Flow, $x=1.63$ m: Solid line is the exponential decay with $K_3=0.7$. 
Figure 26. Lateral \( \gamma \) for the Zero Pressure Gradient Flow, \( x=3.52 \) m: Solid line is the exponential decay with \( K_3=0.7 \).
Figure 27. Lateral $\delta$ for the Zero Pressure Gradient Flow, $x=4.14$ m. Solid line is the exponential decay with $K_3=0.7$. 

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Figure 28. Lateral $y$ for the Zero Pressure Gradient Flow, $x=6.51$ m: Solid line is the exponential decay with $K_3=0.7$. 

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Figure 29. Lateral $\mathbf{y}$ for the Zero Pressure Gradient Flow, $x=6.51$ m ($q=2.4''$H$_2$O), $x=6.51$: solid line is the exponential decay with $K_3=0.7$. 
Figure 30. Longitudinal $\gamma$ for the Favorable Pressure Gradient Flow, $x=1.63$ m: Solid line is the exponential decay with $K_1=0.18$. 
Figure 31. Longitudinal $\gamma$ for the Favorable Pressure Gradient Flow, $x=3.52$ m: Solid line is the exponential decay with $K_1=0.2$. 
Figure 32. Longitudinal $\gamma$ for the Favorable Pressure Gradient Flow, $x=4.14$ m: Solid line is the exponential decay with $K_1=0.18$. 
Figure 33. Longitudinal $\gamma$ for the Favorable Pressure Gradient Flow, $x=4.77$ m: Solid line is the exponential decay with $K_1=0.1$. 
Figure 34. Lateral \( y \) for the Favorable Pressure Gradient Flow, \( x=1.63 \) m: Solid line is the exponential decay with \( k_3=0.35 \).
Figure 35. Lateral $y$ for the Favorable Pressure Gradient Flow, $x=3.52$ m: Solid line is the exponential decay with $K_3=0.5$. 
Figure 36. Lateral $\gamma$ for the Favorable Pressure Gradient Flow, $x=4.14$ m: Solid line is the exponential decay with $K_3=0.6$. 
Figure 37. Lateral $y$ for the Favorable Pressure Gradient Flow, $x=4.77$ m: Solid line is the exponential decay with $K_3=0.6$. 
Figure 38. $U_c/U_e$ versus $\omega_1/U_e$ for the Zero Pressure Gradient Flow: $1.63 \text{ m;}$ $3.52 \text{ m;}$ $4.14 \text{ m;}$ $6.1 \text{ m;}$ $6.51 \text{ m}$ ($q=2.4''\text{H}_2\text{O}$). These are mean curves for all 10 microphone spacings at each x-location.
Figure 39. $U_c/U_e$ versus $\omega \delta_1/U_e$ for the Favorable Pressure Gradient Flow: --- 1.63 m; --- 3.52 m; --- 4.14 m; --- 4.77 m. These are mean curves for all 10 microphone spacings at each x-location.
Table 1. Results of Previous Studies for a Zero Pressure Gradient Flow.
Table 2: Results of Previous Studies for a Favorable Pressure Gradient Flow.

<table>
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<tr>
<th>Authors</th>
<th>$U_e$ m/s</th>
<th>$Re_{X10^{-3}}$</th>
<th>$U/u_e$</th>
<th>$H$</th>
<th>$d^+$</th>
<th>$p'/u_e$</th>
<th>$#(\omega)c^{-1}$</th>
<th>Spectrum level at $u_1/u_e=0.6$</th>
<th>$K_1$</th>
<th>$K_3$</th>
<th>Comments</th>
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<td>BULL (1967)</td>
<td>104</td>
<td>8.0</td>
<td>0.032</td>
<td>1.4</td>
<td>173</td>
<td>2.8</td>
<td>0.4x$\delta$ to $1$</td>
<td>-50 dB</td>
<td>0.1</td>
<td>0.715</td>
<td>Piezoelectric Membrane (Continuous Surface)</td>
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<tr>
<td>SUTERON (1973)</td>
<td>24.38</td>
<td>3.09</td>
<td>0.0488</td>
<td>1.3</td>
<td>60</td>
<td>2.15</td>
<td>No Apparent Region</td>
<td>-66.98 dB</td>
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<td>0.416</td>
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<tr>
<td>DINKELACKER et al. (1977)</td>
<td>8.5</td>
<td>2.0</td>
<td>0.04</td>
<td>1.314</td>
<td>57</td>
<td>NA</td>
<td>$0.026&lt;\omega&lt;0.09$</td>
<td>NA</td>
<td>NA</td>
<td>NA</td>
<td>Optical Device Thin Flexible Membrane</td>
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<tr>
<td>DINKELACKER et al. (1982)</td>
<td>2.9</td>
<td>0.046</td>
<td>0.0517</td>
<td>NA</td>
<td>500</td>
<td>NA</td>
<td>$0.162&lt;\omega&lt;0.568$</td>
<td>NA</td>
<td>NA</td>
<td>NA</td>
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<td>2.0</td>
<td>0.04</td>
<td>1.314</td>
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<td>2.9</td>
<td>No Apparent Region</td>
<td>NA</td>
<td>NA</td>
<td>NA</td>
<td>Optical Device Thin Flexible Membrane</td>
</tr>
<tr>
<td>SCHEWE (1982)</td>
<td>6.3</td>
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<td>19</td>
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<td>$0.242&lt;\omega&lt;0.969$</td>
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<td>NA</td>
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<td>Thin Flexible Membrane Type</td>
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<td>0.925</td>
<td>Flush Mounted Piezoelectric</td>
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* Spectra non-dimensionalized by $10 \log_{10} \left( \frac{\#(\omega)c^{-1}}{U_e} \right)$

NA denotes not available
Table 3. Results of the Present Study for a Zero Pressure Gradient Flows.

<table>
<thead>
<tr>
<th>$X_n$</th>
<th>$U_o$</th>
<th>$H$</th>
<th>$d_{1,mm}$</th>
<th>$U_t$</th>
<th>$Re_9x10^{-3}$</th>
<th>$d^+$</th>
<th>$d^+$</th>
<th>$\frac{p^+}{U_t}$</th>
<th>$\frac{p^+}{U_t}$</th>
<th>$k_3$</th>
<th>$\frac{\delta}{\omega_t/\omega_{10}=1.0}$</th>
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<td>3.329</td>
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<td>79.88</td>
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<td>3.07</td>
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<td>1.38</td>
<td>3.68</td>
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<td>3.818</td>
<td>28.78</td>
<td>79.14</td>
<td>3.82</td>
<td>3.17</td>
<td>----</td>
<td>----</td>
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<td>4.451</td>
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<td>77.58</td>
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<tr>
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<td>5.674</td>
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<td>71.31</td>
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<td>3.79</td>
<td>3.14</td>
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<td>0.7</td>
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* Spectra nondimensionalized by $10 \log_{10} \frac{\delta^+}{\omega_t}$

Comment: Condenser Microphone with Pinhole
Table 4. Results of the Present Study for a Favorable Pressure Gradient Flow.

| X/m | \( u_e/n_0 \) | N | \( 5_1/nm \) | \( U_e/\nu \) | \( Re_\nu x 10^{-3} \) | \( \delta^*_{BT-1753} \) | \( \delta^*_{BT-1755} \) | \( \frac{\rho^*}{\nu} \) | \( \frac{\rho^*}{\nu} \) | \( \frac{u}{U_e} \) | \( \frac{dU_e}{dx} x 10^{-3} \) | \( \psi(u)* \) | Spectrum Level at \( \psi(1)/U_e = 1.0* \) |
|-----|----------------|---|------------|-----------------|-----------------|-----------------|----------------|----------------|----------------|----------------|----------------|----------------|----------------|----------------|
| 1.632 | 11.95 | 1.33 | 3.196 | 0.0454 | 2.447 | 17.64 | 48.52 | 2.618 | 2.873 | 0.18 | 0.35 | 1.31 | |
| 1.882 | 12.63 | 1.32 | 3.419 | 0.0448 | 2.723 | 18.13 | 49.86 | 2.587 | 2.82 | -- | -- | 1.97 | |
| 2.226 | 13.17 | 1.32 | 3.718 | 0.0441 | 3.138 | 18.91 | 52.0 | 2.573 | 2.72 | -- | -- | 1.79 | |
| 2.546 | 13.83 | 1.32 | 3.818 | 0.0438 | 3.384 | 19.71 | 54.19 | 2.574 | 2.69 | -- | -- | 1.84 | |
| 2.958 | 14.52 | 1.31 | 3.915 | 0.0434 | 3.64 | 20.54 | 57.49 | 2.615 | 2.7 | -- | -- | 1.82 | |
| 3.118 | 15.29 | 1.31 | 3.865 | 0.0433 | 3.756 | 21.58 | 59.34 | 2.626 | 2.596 | -- | -- | 2.04 | |
| 3.520 | 16.76 | 1.3 | 3.643 | 0.0434 | 3.912 | 23.49 | 65.16 | 2.611 | 2.516 | 0.2 | 0.5 | 2.11 | |
| 4.146 | 19.56 | 1.28 | 3.141 | 0.0439 | 3.932 | 27.96 | 76.90 | 2.768 | 2.301 | 0.18 | 0.6 | 2.27 | |
| 4.777 | 23.94 | 1.26 | 2.639 | 0.0444 | 6.065 | 34.59 | 95.12 | 2.939 | 2.716 | 0.1 | 0.6 | 2.33 | |

* Spectra nondimensionalized by \( \frac{1}{\psi} \frac{\psi_1}{(\psi_1)^2} \)

Comment: Condenser Microphone with Pinhole
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<tr>
<th>SYMBOLS</th>
<th>STREAMWISE AND SPANWISE SPACINGS $\Delta X$, $\Delta Z$ (mm)</th>
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Table 5. Microphone spacings and symbols for plots of longitudinal and lateral cross spectra for the zero pressure gradient flow.
<table>
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<th>STREAMWISE SPACING $\Delta X$(mm)</th>
<th>SPANWISE SPACING $\Delta Z$(mm)</th>
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Table 6. Microphone spacings and symbols for plots of longitudinal and lateral cross spectra for the favorable pressure gradient flow.
### Abstract

Measurements of surface pressure fluctuation spectra, coherence and convective wave speeds from zero and favorable pressure gradient turbulent boundary layers are reported for momentum Reynolds numbers from 3000 to 18,800. The acceleration parameter, $K$ is near $2 \times 10^{-7}$ for the favorable pressure gradient flow. 

The outer variables, $U_e$, $\tau_w$ and $\delta_l$ non-dimensionalize and collapse the spectra for the low to middle range of frequencies for most test cases. The grouping using the inner variables, $U_\tau$ and $v$, collapse the spectra for the middle to high range of frequencies for all test cases. The value of $p'/\tau_w$ was near 3.8 and 2.8 for the smallest values of $d^+$ in the zero and favorable pressure gradient flows, respectively. 

The coherence exhibits a decay that is not exponential in some cases, but the Corcos similarity parameters $\omega_\delta x/U_c$ and $\omega_\delta z/U_c$ collapse the data for all test cases. The ratio of $U_c/U_e$ increases with $\omega_\delta x/U_e$ up to $\omega_\delta z/U_e$ on the order of unity, where $U_c/U_e$ becomes nearly constant. This was observed in the present results for both streamwise pressure gradient flows. 

The experimental results presented show good agreement with previous research.

### Key Words (Suggested by Authors(s))

- Turbulence
- Boundary layers
- Pressure fluctuations
- Acoustics