MAIN PROPULSION SYSTEM TEST REQUIREMENTS FOR
THE TWO-ENGINE SHUTTLE-C

By E.E. Lynn and G.K. Platt
Propulsion Systems Division
Propulsion Laboratory

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MAIN PROPULSION SYSTEM TEST REQUIREMENTS FOR THE TWO-ENGINE SHUTTLE-C

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February 1990

Delete page 6 and insert new page 6.
**Main Propulsion System Test Requirements for the Two-Engine Shuttle-C**

Prepared by Propulsion Systems Division, Propulsion Laboratory, Science and Engineering Directorate.

The Shuttle-C is an unmanned cargo-carrying derivative of the space shuttle with optional two or three space shuttle main engines (SSME's), whereas the shuttle has three SSME's. Design and operational differences between the Shuttle-C and shuttle were assessed to determine requirements for additional main propulsion system (MPS) verification testing. Also, reviews were made of the shuttle main propulsion test (MPT) program objectives and test results and shuttle flight experience.

It was concluded that, if significant MPS modifications are not made beyond those currently planned, then main propulsion system verification can be concluded with an on-pad propellant loading and countdown demonstration test plus a long duration on-pad flight readiness firing.

**Key Words**

- Shuttle-C
- Propulsion System Testing
- Requirements

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INTRODUCTION

Shuttle-C is a proposed, unmanned, cargo-carrying launch vehicle derived from the space shuttle primarily by replacing the shuttle orbiter with a cargo element and off loading the external tank (ET) propellants. The cargo element will have a much smaller mass than the shuttle orbiter and analyses indicate that Shuttle-C could deliver a target payload of approximately 100,000 lbm to low-Earth orbit with two space shuttle main engines (SSME’s) and approximately 150,000 lbm to low-Earth orbit with three SSME’s. The goal of minimizing development costs makes it desirable to avoid a main propulsion system (MPS) static test program, and the goal of minimizing operating costs makes it desirable to utilize only two SSME’s for payloads in the 100,000-lbm class.

Therefore, a study was requested by the Marshall Space Flight Center (MSFC) Shuttle-C task team to determine whether it would be necessary to conduct a propulsion systems test program to verify the two-engine configuration or whether the new configuration could be verified by other means.

APPROACH USED IN THIS STUDY

In order to determine the need for a Shuttle-C MPS test program, the first step was to conduct detailed reviews of the following:

- Design and operating conditions, Shuttle-C versus shuttle.
- Shuttle MPT program objectives and history.
- Shuttle MPS flight experience.

After these reviews were conducted, a candidate list of MPS related hardware, software, procedures, etc., was generated to identify all areas that could possibly require Shuttle-C MPS test verification. Candidate areas were then eliminated if they could be sufficiently satisfied by the following:

- Shuttle design similarity,
- Shuttle MPT program experience, or
- Shuttle flight experience.
For each area that remained, an assessment was made to determine if it could be adequately verified by any of the following methods:

- Analytical assessment
- Component testing
- On-pad propellant loading/countdown demonstration test(s)
- On-pad flight readiness type hot firing(s)
- Baseline design changes.

**DESIGN AND OPERATING CONDITIONS**

The Shuttle-C reference design is similar to the shuttle’s, except that it is an unmanned but “manrated” vehicle with an expendable propulsion system/payload carrier. The ET/SRB arrangement is the same as that for the shuttle. The Shuttle-C will use two standard SRB’s and a standard ET. However, for a two-engine Shuttle-C the ET will have off-loaded propellants (approximately 80 percent) and relocated liquid level sensors. The two-engine version will have two standard SSME’s with the SSME in position 1 removed and engine interfaces closed off and a heat shield added.

The cargo element to ET interface loads will be within nominal STS design. In flight the Shuttle-C will throttle to limit the load factor to 3 g, but will not throttle to limit the maximum dynamic pressure. There will be limited engine-out capability during the late stages of ascent. Propellant loading procedures, subsystem purges, and engine start preparation procedures will be the same or similar to those used for shuttle.

LO₂ and LH₂ conditions at the engine inlets for engine start do not completely meet Shuttle Interface Control Document requirements. However, based on test histories for engine starts, satisfactory inlet conditions for engine startup are anticipated. The inflight net positive suction pressures at the engine inlets were calculated for flight and meet the requirements.

The Shuttle-C reference configuration is shown in Figure 1. The propulsion subsystem configuration is shown in Figure 2.

**SHUTTLE MAIN PROPULSION TEST PROGRAM**

The primary objectives for a main propulsion system test program are to obtain data on operating characteristics to verify the design of subsystems in a systems environment, to assess the systems operating environment to the extent possible without a flight test, and to detect component deficiencies and defects.
Figure 1. Shuttle-C reference configuration.

- 82' X 15' PAYLOAD BAY
- CIRCULAR P/L CARRIER
- JETTISONABLE SHROUD
- CONICAL NOSE CONE
- MODIFIED ORBITER BOATTAIL
- 2 - 3 SSME's
Figure 2. Shuttle-C propulsion subsystem.
The shuttle main propulsion system test program consisted of 12 MPT firings (see appendix for program history). During these firings, engines were cut off at different times for several reasons, most commonly to make the cut-off transient more conservative or to simulate engine failures. Also, these engine shutdowns were done at different power levels. These test firings helped provide for a better understanding of the propulsion system operating conditions. They also provided the means to verify system prediction capabilities. Table 1 summarizes the MPT one-engine and two-engine operational histories.

During the shuttle MPT program, a number of anomalies and failures were experienced. These are summarized in Table 2. Of these anomalies and failures, all were resolved for the shuttle flight configuration. Many were due to immaturity of components at the time of the system tests. Only those listed below were judged to have required a system test to discover them:

1) LO$_2$ and LH$_2$ prepressurization overshoot
2) LH$_2$ recirculation pump cavitation
3) Nitrogen condensation and LN$_2$ dripping in the engine compartment
4) Heat shield differential pressure exceedances caused by ignition overpressure
5) Inadequate ET nosecap purge
6) LO$_2$ ullage pressure overshoot.

LO$_2$ and LH$_2$ prepressurization were performed prior to test firings and the overshoots were control system problems.

Recirculation pump cavitation was a procedural problem; the pumps were well known, having been used on the S-II and S-IVB stages of the Saturn V launch vehicle.

Nitrogen purge gas commonly condenses on uninsulated or poorly insulated surfaces, and some changes usually have to be made in insulation configuration on any launch vehicle.

Ignition overpressure has been observed in single engine tests. The phenomenon results from the fuel lead during the start transient. When LO$_2$ reaches the injector and ignition begins, an accumulation of hydrogen gas has already occurred in the combustion chamber and nozzle and has become mixed with ambient air. This mixture is then ignited by the combusting propellants. The solution to this problem was to provide igniters or “sparklers” near the nozzle exit to ignite the hydrogen as it mixes with air. In the main propulsion test program, a high overpressure on the orbiter baseplate occurred in one test prior to the inclusion of the external igniters.

The ET nosecap purge problem was one that could have been handled in a component test of the ET; however, the purge verification was planned for MPT instead. The solution was an increase in the nosecap purge gas heater power.
<table>
<thead>
<tr>
<th>Shuttle</th>
<th>E1 Cutoff</th>
<th>E2 Cutoff</th>
<th>E3 Cutoff</th>
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</thead>
<tbody>
<tr>
<td>SF-2</td>
<td>18.8 s (70% power level)</td>
<td>20.6 s (70%)</td>
<td></td>
</tr>
<tr>
<td>SF-4</td>
<td>90 s (70%)</td>
<td></td>
<td>100 s (70%)</td>
</tr>
<tr>
<td>SF-6-04</td>
<td>505 s (70%)</td>
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<td>555 s (70%)</td>
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<tr>
<td>SF-7-02</td>
<td>520 s (70%)</td>
<td></td>
<td>555 s (70%)</td>
</tr>
<tr>
<td>SF-9-02</td>
<td></td>
<td>545 s (65%)</td>
<td>574 s (65%)</td>
</tr>
<tr>
<td>SF-11-02</td>
<td>438 s (65%)</td>
<td></td>
<td>586 s (65%)</td>
</tr>
<tr>
<td>SF-12</td>
<td></td>
<td>235 s (100%)</td>
<td>624 s (100%)</td>
</tr>
</tbody>
</table>

Total 2 engine operational time = 649 s (10.8 min)
Longest 2 engine operational time = 389 s (6.5 min)
Total 1 engine operational time = 29 s (0.5 min)
Longest 1 engine operational time = 29 s (0.5 min)
<table>
<thead>
<tr>
<th>TEST SYSTEM ANOMALIES</th>
<th>ENGINE ANOMALIES</th>
<th>COMPONENT ANOMALIES</th>
<th>PROPULSION SYSTEM ANOMALIES</th>
</tr>
</thead>
<tbody>
<tr>
<td>ET ATTACH BOLTS IMPROPERLY LOADED</td>
<td>MAIN FUEL VALVE RUPTURED</td>
<td>LOX ENGINE CUTOFF LOW-LEVEL SENSORS HAD SLOW RESPONSE</td>
<td>IGNITION OVERPRESSURE</td>
</tr>
<tr>
<td>TEST SYSTEM COMPUTER FAILED</td>
<td>POGO ACCUMULATOR BUBBLE COLLAPSED</td>
<td>GOX PRESSURANT FLOW CONTROL ORIFICES CRACKED</td>
<td>LN₂ IN AFT COMPARTMENT</td>
</tr>
<tr>
<td>ACOSTIC MEASUREMENTS FAILED</td>
<td>ENGINE STEERHORN (NOZZLE HYDROGEN MANIFOLD) RUPTURED</td>
<td>ENGINE LH₂ INLET TEMPERATURE FAILED</td>
<td>LOX TANK NEGATIVE PRESSURE</td>
</tr>
<tr>
<td>ET/ORBITER ATTACH BOLTS OVER TORQUED</td>
<td>FAULTY LOX PREBURNER START</td>
<td>ET/ORBITER DISCONNECT FAILED TO SEAL</td>
<td>ET NOSE CAP PURGE INADEQUATE</td>
</tr>
<tr>
<td>FAULTY HAZARDOUS GAS INDICATIONS</td>
<td>ENGINE HIGH PRESSURE FUEL PUMP LIFTOFF SEAL STUCK</td>
<td>GOX PRESSURANT INTERFACE TEMPERATURE TOO HIGH</td>
<td>LOX TANK ULLAGE PRESSURE OVERSHOOT</td>
</tr>
<tr>
<td>LOX TANK CAPACITANCE PROBE FAILED TO INDICATE</td>
<td>ENGINE HIGH PRESSURE FUEL PUMP SEAL BROKEN</td>
<td>ET HYDROGEN TANK PRESSURANT DIFFUSER FAILURE</td>
<td>LOW LOX TANK ULLAGE PRESSURE DURING RUN</td>
</tr>
<tr>
<td>LH₂ TANK AUXILIARY DRAIN VALVE FAILED TO CLOSE</td>
<td>ENGINE HIGH PRESSURE FUEL PUMP FAILED</td>
<td>IR FIRE DETECTORS FAILED</td>
<td>LO₂ AND LH₂ PREPRESSURIZATION OVERSHOOT</td>
</tr>
<tr>
<td>LOX AUXILIARY DRAIN VALVE STUCK</td>
<td>ENGINE HIGH PRESSURE OXIDIZER TURBINE SEAL FAILED</td>
<td>LOX FILL AND DRAIN VALVE FAILED</td>
<td>LH₂ RECIRCULATION PUMP CAVITATION</td>
</tr>
<tr>
<td>TWO OF 5 H₂ BURNOFF IGNITERS FAILED TO IGNITE</td>
<td>ENGINE FUEL PREBURNER BURN-THROUGH</td>
<td>PROBE SEAL BROKE</td>
<td></td>
</tr>
</tbody>
</table>
The LO₂ ullage pressure overshoot occurred briefly during the early part of firings SF-7-01 and SF-8. This occurred because a vacuum reference system was incorporated to simulate pressure drops with increasing altitude during ascent and the gaseous oxygen (GOX) flow control valves had not been reorificed to compensate for the lower reference pressures for these tests.

It can be seen that all of these anomalies/failures, except ignition overpressure and LO₂ ullage pressure overshoots, were encountered during preparation for firing. Ignition overpressure occurred at engine ignition, and LO₂ ullage pressure overshoot occurred in the early part of firing. Several of these propulsion system problems would have been made worse if one engine had been removed. The indication from these problems experienced during the MPT program is that for the Shuttle-C, critical periods affected by modifications are during propellant conditioning prior to engine start and during the early part of the firing.

SHUTTLE FLIGHT EXPERIENCE

An extensive database of propulsion system related information has been developed through the shuttle flight program that is directly applicable to Shuttle-C main propulsion system verification. In fact, the only propulsion system-related information not applicable at this time are data that would be invalidated by the removal of one SSME, by the off-loading of the propellant tanks, or by the use of foam insulation for the LH₂ feedline instead of using a vacuum-jacketed feedline.

Some two-engine operational experience has been gained through the shuttle flight program. For STS-51F, engine 3 was shutdown at T + 343 s because of an high pressure oxidizer turbopump (HPOTP) discharge temperature redline violation. Engines 1 and 2 continued to operate for an additional 238 s and were cut off from a power level of 91 percent with no anomalous steady-state or shutdown transient characteristics.

All shuttle propulsion system elements have been fully operational since the developmental flight program (STS-1 through STS-4) was concluded. However, some enhancements to performance and reliability are presently being worked. These include the certification of the SSME’s at 109 percent power level, the replacement of the GO₂ flow control valves with orifices, and the possible replacement of the SSME oxidizer heat exchangers with external heat exchangers. These items will be fully verified through subsystem tests and shuttle flight experience. The Shuttle-C program should not require any reverification in order to incorporate these items.

EVALUATION OF CANDIDATE AREAS FOR TEST VERIFICATION

An initial list of areas that could conceivably require verification through Shuttle-C propellant load or hot-fire testing is presented in Table 3. This list was established with major areas presented in a somewhat chronological order. The major areas were broken into applicable subsystems, components, procedures, etc., that might possibly require some Shuttle-C propulsion system test verification.
TABLE 3. INITIAL LIST OF CANDIDATE AREAS FOR SHUTTLE-C PROPULSION SYSTEM TEST VERIFICATION

- Countdown Verification

- Propellant Loading
  - Loading procedures
  - Boiloff and replenish rates
  - Extended countdown hold effects
  - Helium inject system
  - LO₂ bleed/drainback
  - LH₂ recirculation
  - Prepressurization
  - Inlet conditions (start box)
  - Hazardous gas detection system
  - LH₂ high-point bleed

- Buildup Transient Analysis
  - Start sequence for two SSME's
  - Hold down time after SSME ignition
  - Loads transfer through thrust structure
  - Redlines

- Engine Operation
  - Throttling
  - Gimbal angles
    - Angles
    - Rates
  - Nozzle and base heating
  - ME-1 heat shield closeout
  - Net positive suction pressure (NPSP)
  - Loads transfer through thrust structure
  - Engine-out effects
  - Redlines
  - POGO pulsing

- Pressurization System Performance
  - LO₂ ullage pressure slump
  - Flow control operation, nominal
  - Flow control operation with failed flow control valve(s)
  - Pressurization with engine-out

- Propellant Feed System Performance (Surges, Vibrations, etc.)

- Shutdown Analysis
  - Two engine
  - One engine (engine-out case)
  - LO₂ prevalve timing

- Post-Test Checkout and Inspection
Each area was evaluated. Consultation with NASA and contractor engineers from MSFC and Kennedy Space Center (KSC) was obtained in all areas where their experience and expertise could contribute to the identification of test requirements and alternate ways to satisfy them. Results of the evaluations are as follows.

All identified test requirements for the following areas could be satisfied with on-pad propellant loading and short duration FRF tests, similar to the tests conducted at KSC prior to the maiden flight of each space shuttle orbiter:

- Countdown demonstration
- Propellant loading
- Buildup transient analysis
- Shutdown analysis
- Post-test checkout and inspection.

Engine Operation:

- Throttling and Gimbaling — An extensive base of experience has been developed through the shuttle MPT and flight programs. However, Shuttle-C hot-fire verification would be desirable, primarily for SSME gimbaling if profiles differ significantly from those used with shuttle. Combinations of gimbal angles and rates could be tested with an on-pad flight readiness firing (FRF) of sufficient duration.

- Nozzle and Base Heating — The main components of these are radiation and convective heating. Analytical models at MSFC can accurately simulate radiation heating and its effects. Convective heating is more difficult to model. However, convective heating is most pronounced at higher altitudes as the SSME exhaust plumes expand. This condition cannot be duplicated with a sea level hot-firing. Nozzle and base heating are therefore not considered to be drivers for hot-fire testing. However, any hot-fire test that is conducted probably should be done with instrumentation to determine these effects to take advantage of the opportunity.

- ME-1 Heat Shield Closeout — Not a driver for hot-fire testing. However, the opportunity to test for heat transfer to the aft compartment and vibration should be taken on any hot-firing conducted that might reach 1 or 2 min in duration.

- Net Positive Suction Pressure (NPSP) — Can be accurately determined analytically. Any hot-firing would lack solid rocket motor (SRM) and acceleration effects.

- Loads Transfer — This should be verified. Shuttle-C should be instrumented for this. A short duration FRF would be sufficient because the critical loads occur during SSME buildup.
• Engine-out — Shuttle MPT experience with one-engine operation from SF-9-01, SF-9-02, and SF-11-01 is applicable. Therefore, single engine verification is not a driver for hot-fire verification. However, an engine-out conducted during the last stages of an on-pad FRF would provide useful data on the transient effects on various propulsion subsystems and one-engine steady-state performance of the LH₂ and LO₂ pressurization systems.

• Redlines — No requirements for special assessments of performance redlines were identified.

• POGO Pulsing — Extensive POGO testing, conducted during the shuttle MPT program, is directly applicable to Shuttle-C. The SSME POGO accumulators should need no special verification with the Shuttle-C configuration.

Pressurization System Performance:

• LO₂ Ullage Pressure Slump — The pressure drop that occurs shortly after liftoff on shuttle has been determined to be primarily a result of LO₂ tank "breathing" that is caused by SRM thrust buildup loads. The problem is more pronounced because of the small initial LO₂ ullage. Analytical models have been developed that can predict the LO₂ ullage pressure slump.

• Nominal Pressurization — Pressurization system studies indicate that LH₂ tank pressurization performance will not differ enough from shuttle to warrant any special test verification. However, LO₂ pressurization is a different matter. Studies show that with a large initial LO₂ ullage volume and two SSME’s providing GO₂ through standard heat exchangers, flow control valves, and orifices, the ullage pressure control band upper limit of 22 psid will be exceeded at about T + 70 s. The pressure will peak at about 26 psid near T + 100 s and start to decline and will be back inside the control band at about T + 255 s. These LO₂ pressurization studies indicate that the pressurization flow control orifices will have to be resized. An analytical determination of the orifice sizes can be made. However, shuttle pressurization models are based to some extent on empirical shuttle data, and complete confidence in results of resizing the LO₂ flow control orifices would require a relatively lengthy hot-firing.

With everything considered, LO₂ pressurization becomes a prime driver for main propulsion system verification testing. A hot-firing with a duration of approximately 2 min should be adequate to verify LO₂ pressurization system performance.

• Pressurization with Failed Flow Control Valve or Engine-out — These can be determined analytically from nominal pressurization data and require no special testing.

Propellant Feed System Performance:

• No special assessments are required. The LO₂ and LH₂ feedlines to the removed engine will be blocked off flush at the respective manifolds. This should result in no significant change to overall flow dynamics.
MAXIMUM ON-PAD FRF DURATION

Analyses discussed in the preceding section indicate that requirements exist for hot-fire verification for several areas of the Shuttle-C main propulsion system. Two alternatives to conducting these test verifications exist. The first is to conduct an MPT program for Shuttle-C similar to the MPT program that was conducted for shuttle. This utilized a main propulsion test article (MPTA) for a long series of tests at the Stennis Space Center. This would be quite expensive and time consuming. The second alternative, conduct hot-fire test verifications on the pad at KSC, would provide a very significant benefit to the Shuttle-C program in terms of cost and scheduling if all requirements could be adequately satisfied.

On-pad FRF’s of up to 20 s duration have been conducted for shuttle. This duration would be insufficient for Shuttle-C. A study was undertaken to determine the maximum duration that would be feasible for on-pad hot-fire testing at KSC.

The limiting factor on test duration is the amount of water available to satisfy flame deflector cooling and acoustic suppression requirements. The water storage capacities at pads 39A and 39B are 280,000 and 300,000 gallons, respectively. Required flowrates are 65,000 gallons per minute (gpm) for SSME acoustic suppression and 75,000 gpm for flame deflector cooling for a total of 140,000 gpm. For a shuttle launch, there is an additional 400,000 gpm required for solid rocket booster (SRB) related sprays which would be disabled for an extended FRF. Dividing the water capacities by the required flowrates yields a maximum duration of 120 s for pad 39A and 129 s for pad 39B. For an on-pad firing, about 4,000 gallons would have to flow through the water system before flowrates would reach acceptable levels for SSME ignition. This water requirement can be met by the amount initially in the lines below the storage tank bottom. The Firex system utilizes water from a different source and does not affect test duration.

There may be a requirement for heat shielding to protect the ET aft dome area. A heat shield is available for this purpose, although the 20 s shuttle FRF did not utilize the shield and no detrimental results occurred. Also, a study will be required to assure that heat radiated back through the launch pad SRB openings would not be a problem. Shielding could be provided if needed.

SUMMARY

While this study indicates that an MPT program is not required for Shuttle-C, at least one FRF of extended duration will be required. This is only possible because the Shuttle-C is very close to the present shuttle insofar as its propulsion system configuration is concerned. A reexamination of the shuttle MPT program showed that there were many problems uncovered that would have caused extended delays to the shuttle program if they had not been found early through the test article instead of on the first shuttle FRF. Additionally, the ignition overpressure problem experienced in the MPT program could have proven catastrophic if it had occurred in flight.
The primary conclusions of this study are summarized as follows:

1. At least one propellant loading test should be planned. A second test would serve as a contingency if additional verification were required.

2. Hot-fire verification of the main propulsion system will be required.

3. The maximum on-pad FRF duration is approximately 2 min and is limited by the water supply for cooling and acoustic suppression. This matter should be thoroughly evaluated since this greatly exceeds any previous shuttle FRF duration. A water flow test should be conducted to verify the time required to consume the entire water supply.

4. LO₂ tank pressurization has been identified as the most likely area for which a full 2 min duration FRF may be required.

5. It is considered a reasonable goal to verify the Shuttle-C main propulsion system without a separate main propulsion test program, but to rely on propellant loading tests and FRF’s at the launch site. However, consideration should be given to the fact that on-pad FRF testing should be success-oriented. Any anomaly that would require removal of the vehicle from the pad for repairs or modifications could severely impact the program in both cost and scheduling.
APPENDIX

SHUTTLE MPT STATIC FIRING SUMMARY
### MPT Static Firing Summary

**ORGANIZATION:** MARSHALL SPACE FLIGHT CENTER

**DATE:**

<table>
<thead>
<tr>
<th>DATE</th>
<th>TEST NO.</th>
<th>DUR (SEC)</th>
<th>MPS</th>
<th>GIMBAL</th>
<th>POGO</th>
<th>SPECIAL TESTS</th>
<th>POWER PROFILE</th>
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<tr>
<td>4/21/78</td>
<td>15-02</td>
<td>2.35</td>
<td>• Prop. Syst. Shakedown</td>
<td>• SUPPR. OUT</td>
<td>500 Hz Power Supply Evaluation</td>
<td>ICE FROST</td>
<td>3 @ 70% ENG 1 OUT</td>
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<td>5/19/78</td>
<td>2S</td>
<td>20</td>
<td>• Prop. Syst. Shakedown</td>
<td>• SUPPR. OUT</td>
<td>Flare Bucket Eval</td>
<td>Loading Sys Eval</td>
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<td>42</td>
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<td>Pulsing</td>
<td>Suppr. Out</td>
<td>3 @ 90% Sequential Shutdown</td>
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<td>Suppr. Out</td>
<td>2 @ 90%, 1 @ 70% ENG 2 OUT</td>
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</table>

*INTERLOCKS PREVENTED IGNITION ON 6/01 WHICH WAS ATTEMPTED ON 4/11/78*

### MOD Period

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<th>GIMBAL</th>
<th>POGO</th>
<th>SPECIAL TESTS</th>
<th>POWER PROFILE</th>
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<td>5A F</td>
<td>1.5</td>
<td>• Startup Eval</td>
<td>• SUPPR. IN</td>
<td>7 Hour Hold</td>
<td>Loading Systems Eval</td>
<td>5000 GPM LO2 DRAIN</td>
</tr>
<tr>
<td>5/12/79</td>
<td>5F</td>
<td>PLANNED FULL 54</td>
<td>Press Sys Perf</td>
<td>• Freq., • Ramp (130 SEC)</td>
<td>Pulsing 265 SEC</td>
<td>Flare Bucket Eval</td>
<td>1400 GPM LO2 LOAD</td>
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<tr>
<td>7/2/79</td>
<td>6F-01</td>
<td>PLANNED FULL 19</td>
<td>Press Sys Perf</td>
<td>• Step, • Freq., • Ramp (200 SEC)</td>
<td>Pulsing 130 SEC</td>
<td>LO2 Tank Pressure Undershoot</td>
<td>Quick Drain</td>
</tr>
</tbody>
</table>

**REPAIR & MOD PERIOD (UNPLANNED)**
# MPT Static Firing Summary

<table>
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<tr>
<th>DATE</th>
<th>TEST NO.</th>
<th>DUR (SEC)</th>
<th>MPS</th>
<th>GIMBAL</th>
<th>POGO</th>
<th>SPECIAL TESTS</th>
<th>POWER PROFILE</th>
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<tbody>
<tr>
<td>10/24/79</td>
<td>6F-02</td>
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<td>STEP</td>
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<td>FEED SYS PERF</td>
<td>FREQ</td>
<td>SUPPR, IN</td>
<td>1400 GPM FILL</td>
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<tr>
<td></td>
<td></td>
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<td>PNEUM SYS PERF</td>
<td>RAMP</td>
<td>HELIUM BOTTLE BLOWDOWN</td>
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<tr>
<td></td>
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<td>HYD SYS PERF</td>
<td>(250 SEC)</td>
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<td></td>
<td></td>
<td></td>
<td>INTG PROP SYS PERF</td>
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<tr>
<td></td>
<td></td>
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<td>IGNITION OFF</td>
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**REPAIR PERIOD (UNPLANNED)**

**STUB NOZZLES INSTALLED**

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<tr>
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<th>DUR (SEC)</th>
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<th>GIMBAL</th>
<th>POGO</th>
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<tbody>
<tr>
<td>12/17/79</td>
<td>65-04</td>
<td>PLANNED FULL ACTUAL 555</td>
<td>PRESS SYS PERF</td>
<td>STEP</td>
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<td>FEED SYS PERF</td>
<td>FREQ</td>
<td>SUPPR, IN</td>
<td>HELIUM BOTTLE BLOWDOWN</td>
<td>3 AT 100% 3 AT 70% ENG 2 OUT</td>
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<tr>
<td></td>
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<td>PNEUM SYS PERF</td>
<td>RAMP</td>
<td>SEARCH FOR LN2 IN AFT COMPARTMENT</td>
<td>2 @ 90%</td>
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<td>HYD SYS PERF</td>
<td>(250 SEC)</td>
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<td></td>
<td></td>
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<td>INTG PROP SYS PERF</td>
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<tr>
<td></td>
<td></td>
<td></td>
<td>FEED OUT CAL</td>
<td></td>
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<td>SIMULTANEOUS RAMP AND THROTTLE (15 SEC)</td>
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<td></td>
<td>FEED SYS PERF</td>
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<td></td>
<td></td>
<td></td>
<td>OFT-1 SIM</td>
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<td>2 MIN PRESSURIZED HOLD</td>
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<td>LO2 LOW LVL C/O</td>
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<td>PREVALVE RELIEF</td>
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<td>75-02</td>
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<td>SIMULTANEOUS RAMP AND THROTTLE (15 SEC)</td>
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<td>FEED SYS PERF</td>
<td>SUPPR, IN</td>
<td>LO2 TANK PRESSURIZATION OF LO2 TANK</td>
<td>3 AT 100% 3 AT 70% ENG 2 OUT</td>
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</tr>
<tr>
<td></td>
<td></td>
<td></td>
<td>OFT-1 SIM</td>
<td></td>
<td>2 MIN PRESSURIZED HOLD</td>
<td>2 @ 90%</td>
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<td>LO2 LOW LVL C/O</td>
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<td>PREVALVE RELIEF</td>
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<td>INTG PROP SYS PERF</td>
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<td>ICE FROST</td>
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<td></td>
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<td>HYD SYS PERF</td>
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<td>2 @ 90%</td>
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<table>
<thead>
<tr>
<th>DATE</th>
<th>TEST NO.</th>
<th>DUR (SEC)</th>
<th>MPS</th>
<th>GIMBAL</th>
<th>POGO</th>
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<td>3/28/80</td>
<td>85</td>
<td>PLANNED FULL ACTUAL 541</td>
<td>PRESS SYS PERF</td>
<td>RAMP GIMBAL (30 SEC)</td>
<td>SIMULTANEOUS GIMBAL AND POGO (30 SEC)</td>
<td>SENSORS % WET CAL</td>
<td>1 AT 100% 2 AT 70% 3 AT 100%</td>
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<td>FEED SYS PERF</td>
<td>FLIGHT PROFILE (200 SEC)</td>
<td>DWELL (285 SEC)</td>
<td>10MIN SELF PRESSURIZATION OF LH2 TANK</td>
<td>3 AT 100% 3 AT 70% ENG 2 OUT</td>
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<td>DIFFERENTIAL THROTTLING</td>
<td>ATPC CHANNEL FAILURE SIM. (5 SEC)</td>
<td>SUPPR, IN</td>
<td>HYDRAULIC WARMANT LOAD LH2 UNDER PRESS</td>
<td>3 AT 100% 3 AT 70% ENG 2 OUT</td>
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<td>LO2 LOW LVL C/O</td>
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<td>PREVALVE RELIEF</td>
<td>3 AT 100% 3 AT 70% ENG 2 OUT</td>
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<td>3 ENGINE SIMULTANEOUS CUTOFF</td>
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<td>LO2 TANK VENT DOWN WITH ORIFICE</td>
<td>3 AT 100% 3 AT 70% ENG 2 OUT</td>
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<tr>
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<td></td>
<td>ICE FROST</td>
<td>3 AT 100% 3 AT 70% ENG 2 OUT</td>
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## MPT Static Firing Summary

<table>
<thead>
<tr>
<th>DATE</th>
<th>TEST NO.</th>
<th>DUR (SEC)</th>
<th>MPS REQUIREMENTS</th>
<th>GIMBAL</th>
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<tbody>
<tr>
<td>4/15/80</td>
<td>9S-01</td>
<td>PLANNED FULL</td>
<td>O₂ PRESS SYS PERF INTEG. PROP. SYS., PERF.</td>
<td>FREQ (195 SEC)</td>
<td>PULSING (195 SEC)</td>
<td>ICE FROST &amp; IR SCAN, LH₂ RECEIVING MANIFAND AND FEEDLINE RV CHECK, LO₂ FEEDLINE RELIEF, GNC VCV FAILED OPEN, RTLS DUMP, SENSORS CAL, AFT COMPARTMENT, RECIRC PUMP RESTART</td>
<td>ENG 1 OUT, ENG 2 OUT</td>
</tr>
<tr>
<td></td>
<td></td>
<td>ACTUAL 5</td>
<td>FLIGHT LOX LEVEL O₂ LOW LEVEL CUTOFF STS-I CHILLDOWN DWELL THROTTLING</td>
<td>SUPPR. IN</td>
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<td></td>
<td>3 AT 70%</td>
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<tr>
<td>5/30/90</td>
<td>9S-02</td>
<td>PLANNED FULL</td>
<td>O₂ PRESS SYS PERF INTEG. PROP. SYS., PERF.</td>
<td>FREQ (195 SEC)</td>
<td>PULSING (195 SEC)</td>
<td>ICE FROST, LH₂ RECEIVING MANIFAND AND FEEDLINE RV CHECK, LO₂ FEEDLINE RELIEF, GNC VCV FAILED OPEN, RTLS DUMP, SENSORS CAL, LO₂ EOC TEST BY DRAINBACK, FASCOS CHECKOUT</td>
<td>ENG 3 OUT, ENG 2 OUT</td>
</tr>
<tr>
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<td></td>
<td>ACTUAL 574</td>
<td>FLIGHT LOX LEVEL O₂ LOW LEVEL CUTOFF STS-I CHILLDOWN DWELL THROTTLING</td>
<td>SUPPR. IN</td>
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<td>3 AT 63%</td>
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**Mod Period**

Flight Nozzles Installed

<table>
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<th>DATE</th>
<th>TEST NO.</th>
<th>DUR (SEC)</th>
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<th>GIMBAL</th>
<th>POGO</th>
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<tbody>
<tr>
<td>7/12/80</td>
<td>10F</td>
<td>PLANNED FULL</td>
<td>PRESS SYS. PERF ANOMALOUS CONDITION, LO₂ LOW LVL C/O FRF SIMULATION BASE ENV EFFECTS FEEDOUT CAL, FLIGHT CONTROL</td>
<td>NONE</td>
<td></td>
<td>ICE FROST, LH₂ FEEDLINE RELIEF, LH₂ TANK ULLAGE RELIEF, LH₂ TANK ULLAGE PRESSURE CONTROL DURING FILL, GNC FLOW CONTROL VALVE FAILED OPEN</td>
</tr>
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<td></td>
<td></td>
<td>ACTUAL 196</td>
<td>FLIGHT CONTROL</td>
<td></td>
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<tr>
<td>11/3/80</td>
<td>11F-01</td>
<td>PLANNED FULL</td>
<td>PRESS SYS. PERF ANOMALOUS CONDITION, LH₂ LOW LEVEL C/O EMERG THRUST LEVEL EVAL, PUMP OVERSPEED FIX EVAL, FLIGHT CONTROL FEED OUT CAL.</td>
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<tr>
<td></td>
<td></td>
<td>ACTUAL 20 SEC</td>
<td>FREQ, STEP, RAMP, HALTING RAMP (322 SEC)</td>
<td>SUPPR. IN</td>
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</tbody>
</table>

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**Diagram**

- G: Ground<br>
- ENG: Engine<br>
- 1 AT 70%:<br>
- 3 AT 100%:<br>
- 3 AT 66%:<br>
- 3 AT 102%:
# MPT Static Firing Summary

<table>
<thead>
<tr>
<th>DATE</th>
<th>TEST NO.</th>
<th>DUR (SEC)</th>
<th>MPS</th>
<th>GIMBAL</th>
<th>POGO</th>
<th>SPECIAL TESTS</th>
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<tr>
<td>12/4/80</td>
<td>11S-02</td>
<td>PLANNED FULL ACTUAL 586</td>
<td>● PRESS SYS PERF ANOMALOUS COND, ● LH₂ LOW LEVEL CO ● PUMP OVERSPEED AT CUTOFF FIX EVALUATION ● EMERGENCY THRUST LEVEL EVAL.</td>
<td>● HALTING RAMP STEP ● FREQ, RESP ● STEP ● RAMP (290 SEC)</td>
<td>NONE</td>
<td>● ICE FROST ● SHORT THROW IGNITERS FOR GH₂ BURNOFF ● 105 GPM LO₂ BLEED ● GOX FLOW CONTROL VALVE FAILED CLOSED ● LO₂ &amp; LH₂ FEEDLINE RELIEF ● LH₂ DUMP VIA BLEED VALVES</td>
<td>3 AT 100% 3 AT 90% 3 AT 102% 2 AT 90% 2 AT 65%</td>
</tr>
<tr>
<td>12/17/80</td>
<td>12F</td>
<td>PLANNED FULL ACTUAL 624 SEC</td>
<td>● PRESS SYS. PERF., ANOMALOUS COND. ● ANTIGEYSER LINE REMOVED ● EARLY ENGINE OUT ● 5 ENG CUTOFF FROM RPL</td>
<td>● FREQ, RESP, (132 SEC)</td>
<td>PULSING</td>
<td>● ICE FROST ● LONG THROW IGNITERS FOR GH₂ BURNOFF ● GOX &amp; GH₂ FLOW CONTROL VALVES FAILED OPEN</td>
<td>3 AT 102% 2 AT 100%</td>
</tr>
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**MOD PERIOD**

- UPGRADING ENGINES TO FPL CAPABILITY

<table>
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<th>DUR - 515</th>
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<tr>
<td>10/22/81</td>
<td>13F</td>
<td>FULL - 515</td>
<td>● MPS PERF, AT FPL ● BASE ENV, EFFECTS ● LOADS EVAL</td>
<td>TBD</td>
<td>PULSING</td>
<td>● ICE FROST ● ONE LO₂ AND ONE LH₂ FLOW CONTR. VLV FAILED CLOSED</td>
<td>3 AT 90% 2 AT 109% 2 AT 90%</td>
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<table>
<thead>
<tr>
<th>DATE</th>
<th>TEST NO.</th>
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<th>MPS</th>
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<tbody>
<tr>
<td>11/3/81</td>
<td>14F</td>
<td>FULL - 516</td>
<td>● MPS PERF, AT FPL ● TWO ENG CUTOFF FROM RPL ● ONE ENG CUTOFF FROM FPL</td>
<td>● FREQ ● RAMP ● STEP</td>
<td>TBD</td>
<td>TBD</td>
<td>3 AT 90% 2 AT 109% 3 AT 109% 2 AT 90%</td>
</tr>
</tbody>
</table>
MAIN PROPULSION SYSTEM TEST REQUIREMENTS
FOR THE TWO-ENGINE SHUTTLE-C

By E.E. Lynn and G.K. Platt

The information in this report has been reviewed for technical content. Review of any
information concerning Department of Defense or nuclear energy activities or programs has been
made by the MSFC Security Classification Officer. This report, in its entirety, has been determined
to be unclassified.

J.P. McCARTY
Director, Propulsion Laboratory