Seals Flow Code Development

Proceedings of a workshop held at
NASA Lewis Research Center
Cleveland, Ohio
March 26, 1991
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"Numerical/Analytical/Experimental Study of Fluid Dynamic Forces in Seals"

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SUMMARY

In recognition of a deficiency in the current modeling capability for seals, a seven-year effort was established in 1990 by NASA's Office of Aeronautics, Exploration, and Technology under the Earth-to-Orbit Propulsion program of the Civilian Space Technology Initiative to develop verified computational-fluid-dynamic (CFD) concepts, codes, and analyses for seals.

The objectives were to develop advanced concepts for the design and analysis of seals, to effectively disseminate the information to potential users by way of annual workshops, and to provide experimental verification for the models and codes under a wide range of operating conditions. The prime contractor for this effort is Mechanical Technology Incorporated (MTI); the subcontractor is CFD Research Corporation (CFDRC).

The first annual workshop on "Seals Flow Code Development" was conducted at the NASA Lewis Research Center in Cleveland, Ohio, on March 26, 1991. The purpose was to report the progress of this program and to provide a forum for information exchange with the potential end-users. The program accomplishments were presented; discussions were held on the accomplishments and the plans versus the objectives of this program. Workshop participants presented the activities of their respective organizations in the area of seals CFD.

Close to seventy technical managers and designers attended the 1991 seals workshop, including representatives from rocket engine companies, gas turbine companies, seals manufacturers, government agencies, consulting firms and academia. A peer review panel was formed prior to this workshop, representing experts from the user community who are knowledgeable in CFD analysis methods, seal designs, rotordynamics, and/or knowledge-based systems. The function of the peer review panel was to provide recommendations and guidance throughout the program.

The presented material is documented in this NASA Proceedings. The workshop agenda is included in Appendix A; the workshop attendance list is in Appendix B. The scope of the program and the discussions held at the workshop are summarized below.

KNOWLEDGE-BASED SYSTEM ARCHITECTURE

The approach to this program is to develop a knowledge-based system (KBS) for the design and analysis of seals. It involves two major activities: the development of codes for different seal types, and the integration of the various seals and analysis modules into the KBS framework.

The KBS contains and provides the linkage between two interdependent master seal codes: the industrial KBS code and the scientific KBS code. The industrial version is intended to accomplish expeditious analysis, design and optimization of seals; the scientific version is intended to enhance understanding of flow phenomena and mechanisms, to predict performance, to contribute design guidance for complex situations, and to furnish accuracy standards for less sophisticated analyses. Examples of seal types that will be included in the KBS are bushing and ring seals, face seals, labyrinth
seals, tip seals, damping seals, brush seals, electro-fluids seals and "smart" seals.

A master supervisory module (the System Supervisor) in the knowledge-based system provides the integration between the scientific and industrial codes. The interdependence between the two codes is also accomplished by a common database and knowledge-based system utilities, and by using the scientific version as the model which can be used to validate the industrial codes and establish their limits of applicability.

SYSTEM REQUIREMENTS

The hardware and software requirements of the knowledge-based system are dictated by its architectural design. The general objective is to accommodate the various features of the KBS, including seal type and analysis selections, optional program linkage for direct user selection, input modules, output post-processing, design guidance and optimization, and data management. The hardware is required to have sufficient computing capability for these features.

In addition, the system has to include the flexibility for integration, expansion and user access. It has to be designed for future users, anticipating the trends in computer hardware and software development.

The prime contractor, MTI, has devoted a significant effort in the first year of this program to formulate the architecture, define the features and develop the requirements for the system. The approach and the proposed hardware/software configuration were presented and discussed at the 1991 "Seals Flow Code Development" workshop by Dr. Bharat Aggarwal.

The hardware requirements are driven by the need for a graphical user interface as well as by the computational requirements of the scientific codes. The hardware is envisioned to be a desk-top computer or workstation, easily accessible, that will have sufficient memory space for executing the industrial software modules, individually or in a multitasking mode. It will have the capability to allow input/output communications with the scientific codes and to provide the computing power for the use of graphical user interfaces.

The primary hardware options that have been identified are high performance IBM PC compatible computers and Unix-based workstations. The options for a software development environment include: 1. MS-DOS Window 3.0, 2. OS/2 Presentation Manager, and 3. Unix with OSF/Motif Graphical Interface.

The OS/2 Presentation Manager environment has been recommended for software development in this program. This environment, like Unix, offers a powerful operating system and, like Window 3.0, is reasonably priced. The IBM PC compatible high performance computer has been selected as the hardware platform because it has a large installed base with the availability of innovative software development tools. The proposed hardware configuration includes the computer with a minimum of 6 megabyte RAM and 120 megabyte hard disc, Intel 80386 or 80486 processor, and a math coprocessor included for 80386 machines.
Unix-workstation users expressed their concerns about the PC-based choice and the OS/2 operating system as most of the high-technology users are in the process of converting to Unix workstations. The primary issue is whether the selected operating system can be portable across a PC-based computer and a workstation. A second issue is that, if the operating system is to accommodate multiple hardware platforms, similar criterion shall apply to the selection of the software development tools that are going to be used in the operating system.

The requirements and selection criteria that have been developed for the knowledge-based system, as presented at the workshop, provided the basis for the selection of a final system. Based on discussions and feedback from workshop participants, the final selection needs to ensure that it will have the flexibility to satisfy the means of all potential end-users. The program needs to address a wide spectrum of users, from the low-end technology users who do not have the mainframe capability, to the high-technology users who have access to a number of computer systems with unlimited computing power in the network. In addition, the knowledge-based system needs to be modular so that the users can acquire one or multiple modules from either version of the KBS and are able to execute the programs from a computer of their choice.

**SEALS CODE DEMONSTRATION**

Dr. Bharat Aggarwal, Dr. Antonio Artilles and Mr. Wilbur Shapiro of MTI demonstrated the capability of three industrial codes on a 386 computer. The demonstration also included the knowledge-based system (KBS) executive program and the graphical data display capability using current MS-DOS programs. The graphical capability will be ported to OS/2 and made an integral part of the KBS executive program (if OS/2 becomes the choice of operating system for software development).

The three codes demonstrated are for spiral-groove gas seals (SPIRALG), for gas cylindrical seals (GCYL), and for incompressible seals (ICYL). They are modified versions of existing industrial codes. Each code has been implemented on the KBS. The implementation of the code involves the development of the user interface for the knowledge-based system shell (or the executive program for the industrial version of the KBS).

**CFD CODE DEVELOPMENT**

Dr. Andrzej Przekwas of CFD Research Corporation reported the progress of the 3D CFD code for cylindrical seals and the proposed integration of this code into the KBS framework. The aim is to develop a 3D CFD code for the analysis of fluid flow in cylindrical seals and evaluation of the dynamic forces on the seals.

The plans to complete the 3D CFD code for cylindrical seals include modifications to the 3D base code, and incorporation into the 3D code of the physical models to treat turbulence, non-isotropic wall roughness, phase-change and cavitation, a brush-seal model, and the calculation of dynamic coefficients. This code will be demonstrated at the 1992 workshop.

Workshop participants expressed concerns about the users access to the CFD scientific codes. The CFD codes are computation-intensive. They can be best...
executed on a large capacity computer and/or through a network, with input and output communications provided from a desk-top system. Currently, the 3D CFD code is being developed on an Alliant FX-8 computer with 256 megabytes of capacity. The approach is to eventually install the codes on the Cray and provide the necessary linkage between the Cray and the desk-top system. The connection could be local, in the case of high-technology users, or remote, e.g. via a modem or through a network. The procedure for CFD code integration and access will be presented at the 1992 workshop.

INDUSTRIAL INTERESTS

One of the purposes of the annual workshops is to provide interaction between the user community and the code developers, thereby promoting the seal models for practical applications. Close to twenty workshop participants presented seals-related work in their organizations. Other participants expressed interest in supplying test data for code verification and evaluation. Industrial support for this effort was evident and will be pursued as an integral part of this program.

Several major activities are currently in progress in the community in the area of brush seals. The advantage of the brush seal is its potential to reduce leakage. They are being investigated for rocket engine turbopumps, for small engine seals in the air flow systems, and for gas turbine engine applications. Test data are being collected for each application to study the parameters affecting the performance of brush seals. These data can potentially be used to evaluate the brush seal models that will soon be developed in this program.

The industry is continuously interested in the use of the traditional shaft and face seals. Load models and analytical tools are under development for these configurations to improve their performance. Suggestions were made by workshop participants to obtain the analysis of alternate groove patterns to spiral grooves on face seals, to include mechanical face seals, to evaluate the carbon face seal, and to provide means for predicting seal performance in two-phase flow conditions. Buffered seals, rim seals, advanced counter rotating/ultra high pressure noncontacting seals, and gas seals for 3600 rpm range were also among those recommended by workshop participants to be included in model evaluation.

One of the key items on the agenda for the 1992 workshop will be the issue of technology transfer. The users are anxious to acquire the codes that are developed in this program, and exercise them on their own designs and problems. Prompt technology transfer will provide benefits to the program in two ways: 1. the codes will be evaluated by actual test data over practical operating conditions and become more valuable, and 2. the users will have enhanced analytical capabilities and develop confidence in these codes for the analysis and design of seals.

Typically, NASA sponsored software tools are released via COSMIC, the Computer Software Management and Information Center operated by the University of Georgia for the Technology Utilization Office. In this program, the same mechanism will be used for the final release of the completed codes. In the interim, a technology transfer plan will be developed prior to the 1992 workshop to define a transfer-and-feedback mechanism for these codes so that
they can be released promptly to the user community for evaluation and
verification prior to final delivery to COSMIC. The codes developed under
this program are targeted for a wide user community. Through an effective
technology transfer mechanism, they can be calibrated and become useful tools
for commercial use.

Anita D. Liang
Workshop Chairman
INTRODUCTION

Anita D. Liang
NASA Lewis Research Center

NASA Contract NAS3-25644, "Numerical/Analytical/Experimental Study of Fluid Dynamic Forces in Seals", was initiated in 1990 by the NASA Lewis Research Center. It is a seven year program; the Mechanical Technology, Inc. (MTI) is the prime contractor and CFDRC is the subcontractor. Annual workshops are included as part of the program to provide the interaction between the user community and the contract team, and to disseminate the codes that are developed in this program to the potential end-users.

An introduction is made at the beginning of the first annual "Seals Flow Code Development" workshop to give the participants an overall description of the program. The program scope, the deliverables, the first-year accomplishments and the near term plans are presented.

- Description of Contract, NAS3-25644
  Numerical/Analytical/Experimental Study
  of Fluid Dynamic Forces in Seals
- Program Status, Accomplishments and Plans
- Workshop Objective
TEAM MEMBERS

NASA LeRC
Bob Hendricks
Anita Liang
Margaret Proctor
Julie Schlumberger

MTI
Bharat Aggarwal
Tony Artiles
Wilbur Shapiro

CFDRC
Mahesh Athavale
Andrzej Przekwas
Ashok Singhal

PROGRAM OBJECTIVE - Develop Codes for Analyzing and Designing Optimized Advanced Seals for Future Aerospace and Advanced Rocket Engine Systems

- DEVELOP 3-D CFD SCIENTIFIC CODES
- COMPILIE AND GENERATE SETS OF VERIFIED 2-D INDUSTRIAL CODES
- GENERATE A KNOWLEDGE BASED SYSTEM (KBS) TO BE COUPLED TO THE CODES
**PROGRAM OUTLINE**

- **SEVEN YEAR EFFORT**

- **PARALLEL PATHS**
  - SCIENTIFIC CODE WILL CONTRIBUTE TO TECHNICAL DATA BASE AND PROVIDE AN ACCURACY STANDARD FOR LESS COMPLEX CODES
  - INDUSTRIAL CODES WILL PROVIDE SPEEDY ANALYSIS OF SEAL DESIGNS

- **KBS IS UNIFYING MECHANISM**

- **TECHNOLOGY TRANSFER WILL BE ACCOMPLISHED VIA WORKSHOPS**

- **PROGRAM WORKPLAN WILL BE REVIEWED AND REVISED ANNUALLY**

**PROGRAM SCHEDULE**

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<thead>
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<td>Task IV: Face, Wave, &amp; Grooved Seals</td>
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<td>Task V: Tip Contact &amp; Non-continuous seals</td>
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<td>Task VI: Code Verification and Design</td>
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<td>Task VII: Design Optimization of KBS Seals</td>
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<td>Task VIII: Turbomachinery</td>
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<td>Task IX: Test Data Feedback</td>
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<td>Task X: Development of KBS Code (Scientific)</td>
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<td>Task XI: Development of KBS Code (Industrial)</td>
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<td>Task XII: Reports &amp; Workshops</td>
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</table>
COMMON SCOPE OF CODE DEVELOPMENT TASKS
(TASKS I THROUGH IV, AND TASK IX)

- LITERATURE SURVEY
- CODE FORMULATION
- OUTLINE FOR KBS INTEGRATION
- PROGRAMMING AND CODING
- VERIFICATION OF THE NUMERICAL ANALYSIS
- CODE DOCUMENTATION
- CODE INSTALLATION AND TECHNOLOGY TRANSFER

TASK V - CODE VERIFICATION AND DESIGN

CODE VERIFICATION

1. PUBLISHED LITERATURE
2. UNPUBLISHED INFORMATION FROM INDUSTRIAL SOURCES
3. CONSULTATION

DESIGN OF KBS CODES
PROGRAM ORGANIZATION

MTI PROGRAM MANAGER
W. SHAPIRO

MTI PRINCIPALS
A. ARTILES
B. AGGARWAL

NASA LERC
A. LIANG
R. HENDRICKS

PEER REVIEW PANEL

CFDRC
A. PRZEKWAS
M. ATHAVALE

TASK V
CODE VERIFICATION AND
KBS DESIGN

TASK VI
DESIGN OPTIMIZATION

TASK VII
TURBOMACHINERY
TEST DATA FEEDBACK

TASK VIII
KBS SCIENTIFIC
CODE

TASK IX
KBS INDUSTRIAL
CODE

Task 1 Cylindrical Seals Code
Task II Cylindrical Code Augmentation (Labyrinth, Honeycomb, Damper, Brush)
Task III Face, Wave, Grooved Seals
Task IV Tip, Contact, Non-Continuous Seals

CODE DELIVERABLES

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<tr>
<th>CODE OR MODULE</th>
<th>APPROXIMATE DELIVERY DATE</th>
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<td>1. CFD Cylindrical Code</td>
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<td>2. Augmented CFD Cylindrical Module</td>
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<td>3. CFD Code, Face, Wave, Non-Continuous Module</td>
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<td>4. CFD, Tip, Contact, Non-Continuous Module</td>
<td>09/01/96</td>
</tr>
<tr>
<td>5. GJOURN</td>
<td>02/01/91</td>
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<td>6. ICYL</td>
<td>03/01/91</td>
</tr>
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<td>7. SPIRALG</td>
<td>04/01/91</td>
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<td>8. IFACE</td>
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<tr>
<td>9. GFACE</td>
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<td>10. SPIRALI</td>
<td>04/01/92</td>
</tr>
<tr>
<td>11. FACEDY</td>
<td>02/01/93</td>
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<tr>
<td>12. RINGDY</td>
<td>03/01/93</td>
</tr>
<tr>
<td>13. LABYRINTH</td>
<td>04/01/93</td>
</tr>
<tr>
<td>14. FACECON</td>
<td>02/01/94</td>
</tr>
<tr>
<td>15. DISTORTION</td>
<td>04/01/94</td>
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<tr>
<td>16. Additional Codes</td>
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<td>17. Industrial KBS</td>
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<td>18. Scientific KBS</td>
<td>04/01/96</td>
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ACCOMPLISHMENTS

- APPROVED THE BASE WORKPLAN
- COMPLETED 3 INDUSTRIAL CODES
  - ICYL
  - GCYL
  - SPIRALG (SPIRALGC + SPIRALGF)
- COMPLETED FORMULATION OF THE KBS
  STARTED INTEGRATION OF THE 3 CODES
- DEVELOPED NEW AND MORE EFFICIENT ALGORITHMS
  FOR CFD ANALYSIS
- FORMED THE PEER REVIEW PANEL
- ORGANIZED THE 1ST ANNUAL WORKSHOP

SEALS CODE USER COMMUNITY

ROCKET ENGINE COMPANIES
GAS TURBINE MANUFACTURERS
SEAL COMPANIES
CONSULTANTS
GOVERNMENT AGENCIES
UNIVERSITIES
PEER REVIEW PANEL

GEORGE BACHE - AEROJET
TED BENJAMIN - NASA MSFC
DARA CHILDS/GERALD MORRISON - TEXAS A&M UNIVERSITY
RONALD DAYTON - U.S. AIR FORCE
HAROLD GREINER - EG&G
EUGENE JACKSON/JOE SCHARRER - ROCKWELL
ALAN LEBECK - MECHANICAL SEAL TECHNOLOGY
CHARLES MERKLE - PENN STATE UNIVERSITY
JOHN MINER/SAADAT SYED - PRATT & WHITNEY
JOHN MUNSON - ALLISON GAS TURBINE DIVISION
NELSON POPE - GENERAL ELECTRIC
RICHARD SALANT - GEORGIA INSTITUTE OF TECHNOLOGY

WORKSHOP RESULTS

- NASA CONFERENCE PROCEEDINGS
  - WORKSHOP PRESENTATION MATERIAL
  - EXECUTIVE SUMMARY

- UPDATED WORKPLAN
  - QUESTIONNAIRE
  - PEER REVIEW RECOMMENDATIONS
  - WORKSHOP DISCUSSIONS
NEXT YEAR’S WORKSHOP

- UPDATED WORKPLAN
- CYLINDRICAL CFD CODE
- IFACE, GFACE, SPIRAL INDUSTRIAL CODES
- FEEDBACK ON THE FIRST THREE CODES DELIVERED IN 1991
- TECHNOLOGY TRANSFER AND UTILIZATION PLAN
- PRESENTATIONS BY PARTICIPANTS
The industrial codes will consist of modules of 2-D and simplified 3-D or 1-D codes, intended for expeditious parametric studies, analysis and design of a wide variety of seals. Integration into a unified system is accomplished by the industrial KBS, which will also provide User Friendly interaction, contact sensitive and hypertext help, design guidance and an expandable database.

Major seal types include Cylindrical Seals and Face Seals, and within each type are a wide variety of configurations. For example, cylindrical seals include uniform, step, taper, hydrostatic, segmented, damping, labyrinth, and spiral-groove configurations. More advanced types include brush seals, electro-fluids seals and smart seals.

The types of analyses to be included with the industrial codes are interfacial performance (leakage, load, stiffness, friction losses, etc.), thermoelastic distortions, and dynamic response to rotor excursions.

The first three codes to be completed and which are presently being incorporated into the KBS are the compressible spiral-groove codes, SPIRALG, the incompressible cylindrical code, ICYL, and the compressible cylindrical code, GCYL.

The spiral-groove codes analyzes both shaft seals and face seals with finite eccentricity and misalignment. Four degrees of freedom are included for cylindrical seals, and three for face seals. The code predicts load, flow, power loss, and cross-coupled dynamic spring and damping coefficients, shaft displacements and minimum film thickness.

The ICYL (Incompressible Cylindrical Code) is a 2-D isoviscous code that includes roughness, multiple geometries (steps, pockets, tapers, preloaded axes, and hydrostatic), turbulence, cavitation, and inertia at inlets to the film. Included are eccentricity and misalignment and variable grid specifications. Specified pressure and periodic boundary conditions can be applied. It produces pressures, flows, load, righting moments, film thickness, power loss and cross-coupled dynamic spring and damping coefficients.

GCYL is a gas cylindrical code that can treat varying geometries (steps, tapers, hydrostatic, lobed, segmented). Shaft eccentricity, misalignment, specified boundary pressures or periodic boundary conditions can be applied. The program produces the clearance distribution, pressure distribution, leakage, load, moments, power loss and cross-coupled dynamic spring and damping coefficients.
Objectives:

• Compile and generate sets of verified 2D and simplified 3D or 1D codes

• Codes are intended for expeditious parametric studies, analysis and design of a wide variety of seals

• Integration is accomplished by the Industrial KBS.

Additional functions of the KBS are:

- User Friendly Interaction
- Contact Sensitive and Hypertext Help
- Design Guidance
- Expandable Database

SEAL TYPES - COMPRESSIBLE OR INCOMPRESSIBLE

Bushing and Ring Seals

• Uniform
• Axial Step
• Axial Taper
• Hydrodynamic Step
• Hydrodynamic Taper
• Self-Energized Hydrostatic
• Segmented
• Damping Seals
• Spiral Groove
Circumferential Multilobe
(with or without grooves)

Tapered in Flow Direction

Self-Energized —
Hydrostatic
(Inherent Compensation
Orifice Compensation
Spot Orifices
Recesses)

Circumferential Rayleigh Step

Rayleigh Step in Direction of Flow

SEGMENTED RING SEAL

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<th>Item</th>
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<td>Segmented Ring</td>
<td>Carbon</td>
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<td>2</td>
<td>Rayleigh Step</td>
<td>Inconel X-750</td>
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<td>3</td>
<td>Spring-Back</td>
<td>Inconel X-750</td>
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<td>4</td>
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<td>5</td>
<td>Cover</td>
<td>Stainless Steel 17-4 FH</td>
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<td>6</td>
<td>Step Pin</td>
<td>Stainless Steel 17-4 FH</td>
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<tr>
<td>7</td>
<td>Seal</td>
<td>Teflon</td>
</tr>
<tr>
<td>8</td>
<td>Sleeve</td>
<td>Inconel 718 Hard Chromium Plated</td>
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</table>

(ORIGINAL PAGE IS OF POOR QUALITY)
DAMPING SEAL-LEAKAGE FLOW DAMPENS ROTOR MOTION;
ROUGH STATOR HINDERS ROTOR WHIRL

SPIRAL-GROOVE
SEAL TYPES – COMPRESSIBLE OR INCOMPRESSIBLE

Face Seals

- Contact Face Seals
- Radial Step
- Radial Taper
- Hydrodynamic Step
- Hydrodynamic Taper
- Hydrostatic
- Spiral Groove
- Multi-pad

Hydrostatic

Circumferential Rayleigh Step

Circumferential Tapered Land

Radial Rayleigh Step

Radial Tapered Land

GFACE Configurations
Direction of Rotation
Providing Pumping
Grooves Rotate

Land Width
Groove Width

Groove Angle = $\alpha$
Groove Depth = GD
Land Width/Groove Width = $\gamma$

SPIRAL GROOVE PARAMETERS
CONTACT FACE SEAL WITH ROTATING SEAL RING

CONTACT FACE SEAL WITH NON-ROTATING SEAL RING

(Dimensions in mm Unless Otherwise Noted)
Brush Seals

Labyrinth Seals

• Straight
• Stepped
• Abradable
• Angled

Electro-Fluids Seals

Smart Seals

Effect of Bristle Overhang

Small overhang gives small axial deflection

Larger overhang gives larger axial deflection

Position of rotor CD

X Section A-A

DIAGRAM SHOWING TYPICAL BRUSH SEAL ARRANGEMENT
"C" Radial Clearance

ORIGINAL INLET SEAL

Dimensions in inches unless otherwise specified

881815

"C" Radial Clearance

OPTIMIZED STRAIGHT INLET SEAL

Dimensions in inches unless otherwise specified

881816

23
MAGNETIC FLUID SEAL

HIGH-SPEED POSITION OF FLUID (CENTRIFUGAL SEAL)

HEAT EXCHANGER
COOLING WATER

ZERO OR SLOW SPEED POSITION OF FLUID (MAGNETIC SEAL)

NONMAGNETIC MATERIAL

MAGNETIC MATERIAL, MAGNETIZED BY PERMANENT OR ELECTROMAGNET

HIGH-PRESSURE REGION

LOW-PRESSURE REGION

ROTATING COLLAR

FUNDAMENTAL SEAL CONFIGURATION

Optimize Geometry for Seal Performance

Dynamic Response

Determine if Dynamic Response is Acceptable

Interfacial Analysis

Thermoelectric Distortions

Determine if Distortions are Excessive

ANALYTICAL TRIAD
Code Deliverables

<table>
<thead>
<tr>
<th>Code Module</th>
<th>Approximate Delivery Date</th>
</tr>
</thead>
<tbody>
<tr>
<td>GCYL (Gas Cylindrical)</td>
<td>02/01/91</td>
</tr>
<tr>
<td>ICYL (Incompressible Cylindrical)</td>
<td>03/01/91</td>
</tr>
<tr>
<td>SPIRALG (Gas Spiral Groove)</td>
<td>04/01/91</td>
</tr>
<tr>
<td>CFD Cylindrical Code</td>
<td>02/01/92</td>
</tr>
<tr>
<td>IFACE (Incompressible Face)</td>
<td>02/01/92</td>
</tr>
<tr>
<td>GFACE (Gas Face)</td>
<td>03/01/92</td>
</tr>
<tr>
<td>SPIRALI (Incompressible-Spiral Groove)</td>
<td>04/01/92</td>
</tr>
<tr>
<td>FACEDY (Face-Dynamics)</td>
<td>02/01/93</td>
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<tr>
<td>RINGDY (Ring-Dynamics)</td>
<td>03/01/93</td>
</tr>
<tr>
<td>Labyrinth (Gas)</td>
<td>04/01/93</td>
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<tr>
<td>Augmented CFD Cylindrical Module</td>
<td>02/01/94</td>
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<tr>
<td>FACECON (Face Contact)</td>
<td>02/01/94</td>
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<tr>
<td>DISTORTION (Thermo Elastic Distortion)</td>
<td>04/01/94</td>
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<tr>
<td>Additional Codes</td>
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<tr>
<td>- Brush</td>
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<tr>
<td>- Damping Seal</td>
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<tr>
<td>CFD Code, Face, Wave, Groove Module</td>
<td>09/01/95</td>
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<td>Industrial KBS</td>
<td>04/01/96</td>
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<tr>
<td>Scientific KBS</td>
<td>04/01/96</td>
</tr>
<tr>
<td>CFD, Tip, Contact, Non-Continuous Module</td>
<td>09/01/96</td>
</tr>
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</table>
PLANNED ACTIVITY

- Code unification and completion
  - Unified Grid Generator
  - On-line Help

- Code documentation
  - Consistent with KBS usage

- Additional problems and checkout

- Annual report
This presentation is a status report for Task I of the Seals Flow Code project. The aim of this task is to develop a 3D CFD code for the analysis of fluid flow in cylindrical seals and evaluation of the dynamic forces on the seals. This code is expected to serve as a scientific tool for detailed flow analysis as well as a check for the accuracy of the 2D industrial codes.

The features necessary in the CFD code are outlined. These include general considerations such as code accuracy, efficiency, and robustness and other specific features such as rotating coordinate frames, steady and time-accurate solutions, inclusion of non-isotropic wall roughness, effects of phase-change and cavitation, and integration with KBS.

The approach and initial focus of the task is discussed next. The initial focus of the task was to develop or modify and implement new techniques and physical models. These include collocated grid formulation, rotating coordinate frames and moving grid formulation. Other advanced numerical techniques include higher-order spatial and temporal differencing and efficient linear equation solver.

These techniques were implemented in a 2D flow solver for initial testing. Several benchmark test cases were computed using the 2D code and the results of these are compared with analytical solutions or experimental data to check the accuracy. Tests presented here include planar wedge flow, flow due to an enclosed rotor and flow in a 2D seal with a whirling rotor. Comparisons between numerical and experimental results for an annular seal and a 7 cavity labyrinth seal are also included.

The presentation is closed with plans for the second and final year of Task I. This involves completion of modifications in the 3D base code, and incorporation in the 3D code of the physical models to treat turbulence, non-isotropic wall roughness, phase-change and cavitation, brush-seal model, and calculation of dynamic coefficients.
OBJECTIVES

• Develop Verified CFD Code for Analyzing Seals

• Required Features Include:
  • Applicability to a Wide Variety of Seal Configurations, such as: Cylindrical, Labyrinth, Face, Tip and Brush Seals
  • Accuracy of Predicted Flow Fields and Dynamic Forces
  • Efficiency (Economy) of Numerical Solutions
  • Reliability (Verification) of Solutions
  • Ease-of-Use of the Code (Documentation, Training)
  • Integration with KBS

CFD CODE REQUIRED CAPABILITIES

• Solve 2-D and 3-D Navier-Stokes Equations with Provisions for:
  • Complex Geometries (Body Fitted Coordinates, High Aspect Ratio Cells, Skewed Grids, etc.)
  • Stationary and Rotating Coordinates
  • Steady-State and Transient Analysis
  • Turbulence, Surface Roughness
  • Compressibility, Viscous Dissipation
  • Heat Transfer, Phase Change, Cavitation
  • Electromagnetic Effects, etc.
CFD SEALS CODE DEVELOP. - APPROACH

- Use state-of-the art techniques and latest relevant physical models
- Starting Point: REFLEQS-2D and 3D codes
- 2D code used as a base for efficient evaluation of latest numerical techniques
- Optimum procedure and programming practice tested in 2D code (computer memory vs. computational speed dilemma)
- Fast turnaround time
- Several benchmark-quality validation experiments available for 2D and only few for 3D
- Verified 2D code used as a checkout test bed for 3D code

INITIAL FOCUS OF TASK 1

- Study, Implement and Test:
  a) Non-Staggered Grid Formulation
  b) Rotating Coordinates
  c) Transient Solution Algorithm

- Physical Model Modifications for:
  a) Turbulence in Rotating Flows
  b) Non-Isotropic Wall Roughness
  c) Compressibility and Viscous Dissipation
  d) Two-Phase Flows (Phase Cavitation, etc.)

- Calculation of Dynamic Forces and Coefficients
ADVANCED NUMERICAL TECHNIQUES - STATUS

- Colocated Grid Formulation for BFC Grids
- Strong Conservative Formulation of Momentum Equations with Cartesian Components
- Choice of SIMPLE, Modified SIMPLEC and Noniterative PISO
- Higher Order Accurate Temporal Differencing
- Higher Order (2nd, 3rd) Spatial Discretizations Available
- Rotating Grid System for Stator-Rotor Configurations
- Moving Grid option for Arbitrary Rotor Whirling Analysis
- Efficient Multigrid Solver

PLANAR WEDGE FLOW

Length = 0.1, Heights: Minimum = 3 \times 10^5 m, Maximum = 3 \times 10^{-4} m, y scale enlarged 200 times

192 x 40 Grid

Streamlines

\[ \text{Streamlines} \]

\[ \text{U = 15 m/s} \]
PLANAR WEDGE FLOW

u Velocity

Pressure

ENCLOSED ROTOR

Daily & Nece (1960)
ENCLOSED ROTOR
Daily and Nece

Tangential Velocity

Radial Velocity

Data of Daily & Nece
Colocated Grid

Data of Daily & Nece
Colocated Grid

WHIRLING ROTOR IN AN ANNULAR SEAL

Stator radius = 50.1478 x 10^{-3} m,
Clearance = 495 \mu m, \varepsilon = 0.8 \times clearance,
Shaft Speed \omega = 5085 \text{ rpm}

40 x 10 Computational Grid

Pressure Contours
Synchronous Whirl, \Omega = \omega

ORIGINAL PAGE IS
OF POOR QUALITY
WHIRLING ROTOR IN AN ANNULAR SEAL

Pressure Contours
Subsynchronous Whirl, $\Omega = 0.5\omega$

Pressure Contours
Subsynchronous Whirl, $\Omega = 0.01\omega$

ANNULAR SEAL

Morrison, et al.
$Re = 27000, \; T_a = 6600, \; L/C = 29.4$

Annular Rotor

Seal Geometry

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ANNULAR SEAL
$u_x/\bar{U}$ contours

Experimental

Numerical

Inlet

Outlet

SEVEN CAVITY LABYRINTH SEAL

Morrison, et. al.
Re = 28000  Ta = 7000

Labyrinth Rotor

Inlet Geometry

ORIGINAL PAGE IS OF POOR QUALITY
SEVEN CAVITY LABYRINTH SEAL
Velocity Vector Plot

Experimental       Numerical

First Cavity

Third Cavity
SEVEN CAVITY LABYRINTH SEAL
$u_x/\bar{U}$ contours

Experimental  Numerical

First Cavity

Third Cavity
SEVEN CAVITY LABYRINTH SEAL
\( u_r/\bar{U} \) contours

First Cavity

Third Cavity
SEVEN CAVITY LABYRINTH SEAL

$\frac{u_0}{w_{shaft}}$ contours

Experimental | Numerical

First Cavity

Third Cavity
TASK 1 COMPLETION PLAN (DEC 91)

- Complete Advanced Numerical Techniques Implementation
- 3D Code Modifications
- Incorporation of Advanced Physical Models
  - Turbulence Models
  - Non-Isotropic Wall Roughness
  - Compressibility and Viscous Dissipation
  - Two-Phase (Phase Change)
  - Cavitation
  - Brush Seal Model
- Incorporation of Dynamic Forces and Coefficients
KBS DEVELOPMENT

Bharat Aggarwal
Mechanical Technology Inc.

The KBS has two main objectives: 1) Provide easy access to all the industrial and scientific codes through an executive program. The executive provides common services such as browsing and printing, direct access to a specific code by clicking a button or selecting from a menu, and expert help and guidance for seal analysis and design using expert systems to assist the user. Other features of the KBS include online help and access to a database of test cases for the analysis codes. 2) Provide an easy to use user interface for the analysis codes. The user interface is designed to minimize the effort expended in dealing with the mechanics of using a computer, allowing the user to concentrate on seal design and analysis.

The KBS is implemented using the OS/2 Operating System with the Presentation Manager (PM) graphical user interface. The PM environment provides support for windows, menus, buttons, etc. to enable design of a friendly user interface. In addition, a rich set of functions are provided to support the development of interactive graphics applications. This facility will be used to provide capabilities such as interactive seal layout and grid generation to reduce the volume of numeric input required by the programs. OS/2 currently runs on IBM PC compatibles using Intel 80286, 80386, or 80486 processors. IBM and Microsoft plan to port the system to run on some of the RISC platforms by the end of 1992. The OS/2 Extended Edition from IBM also provides integrated communications and database support. These facilities will be used extensively when the scientific codes are integrated into the KBS.

The user interface is designed to reduce the mechanics of using a computer to simple operations such as menu selections, clicking buttons, etc. The basic operations and menu items are consistent for all the programs to reduce the learning time and to make the user feel comfortable when using the various programs. Context sensitive help is provided at all times and the user is protected from making obvious errors. The data entry procedures are designed using engineering terminology rather than computer jargon. The interactive portions of the input are being designed to conform to the level of abstraction used in the theoretical model so that the user does not waste time and effort translating information on the screen into the concepts used to build the analytical model. The end result will be to have the interface recede into the background, leaving the user free to concentrate on seal design and analysis.
Objectives

- Develop 3-D CFD Scientific Codes (CFDRC)
  - Highly accurate
  - Standards for 2-D codes
- Generate 2-D Industrial Codes
  - Expeditious analysis
- Develop Knowledge-based System (KBS)
  - Integrate all analysis codes and modules
  - Design guidance
  - An upgradable database
  - User friendly input and output procedures

Hardware and Software Configuration

- IBM PS/2 Model 80-A31
  - 25 MHz 80386 processor
  - 25 MHz 80387 math coprocessor
  - 8 Mb RAM
  - 320 Mb high speed hard disk
  - 1.44 Mb 3.5 inch floppy drive
  - SCSI adapter with cache
- IBM High Resolution (VGA) Color Monitor
- Microsoft Serial PS/2 Mouse
- IBM OS/2 Extended Edition (V 1.2)
  - OS/2 Standard Edition
  - Communication Manager
  - Database Manager
- IBM OS/2 Programming Tools and Information Kit (V 1.2)
- Microsoft C Compiler (V 6)
- Microsoft FORTRAN (V 5)
- Kedit for OS/2
- Asymmetrix Toolbook for OS/2
- Nexpert Object for OS/2
O Design Philosophy for KBS

- A consistent user interface
  - Reduce learning time and make the user feel at ease
  - Appearance of windows, input screens, etc. will be the same in all modules
  - All modules will use similar procedures for standard operations such as opening or closing files, printing reports, selecting items from menus, etc.
  - Names of menu items will be standardized as much as possible
  - Context sensitive help available at every stage
  - Provide facilities to recover from errors

- The user has complete control
  - The user can perform functions in any order
  - The program will warn user if the information needed to do the job is not available

O KBS: Analysis Definition

- Analysis Types
  - Interfacial (2-D and 3-D)
  - Dynamic response
  - Thermoelastic distortion

- Direct Selection by the User
  - User selects analysis type from a menu

- KBS Assistance
  - User provides information relative to the application
  - A knowledge base to analyze the information and recommend analysis type and the analysis codes to be used
KBS: Design Guidance and Optimization

Guide iterative analysis

- Optimization is an iterative process involving interfacial analysis, dynamic analysis, and thermoelastic analysis
- Analyze output data to determine if iterations should continue
- Recommend the analysis to run in the next iteration
- Recommend the parameters to be changed in the next iteration

KBS: Databases

Data to be Archived

- User Input data sets
- User Output data sets
- Data sets used in code validation

Database Procedures

- OS/2 Extended Edition Database Manager used for storing and retrieving data
- Networked database support provided through OS/2 facilities
- Data entry and retrieval screens will be designed using the Query Manager and SQL compatible commands
- Password protection will be provided to prevent unauthorized access
KBS: Help and Guidance

Implementation

- Hypertext capabilities of Toolbook and the Information Presentation Facility (IPF) included with OS/2 will be used to design the help system
- Graphics and text will be mixed as needed
- Context sensitive help

Help: General

- User Interface

Help: Each Analysis Module

- Purpose
- Capabilities and limitations
- Theoretical background
- References
- Code validation procedures
- Description of input and output variables
- Example problems with sample input and output

Guidance

- Design procedures
- Seal type recommendations

KBS: Menu Structure

CFD Codes - Main Menu
Scientific Codes
Industrial Codes
Analysis Definition
Seal Selection
Database Operations
Design Guidance
Help
Quit
Output File:

Input File:

CFD Industrial Codes: Spiral Groove Gas Seals

**Grid Definition**

- Seal Configuration
  - **1 Region**
  - **2 Regions**
  - **3 Regions**

- Number of Circumferential Grid Intervals
- Relative Size
- Grid Intervals
- Groove/Pitch Ratio
- Groove Angle (Deg)
- Groove Depth (in)
The demonstration included five different items; the executive program, three of the industrial codes, and graphical data display capability using current MTI MS-DOS programs. The graphics capability included two- and three-dimensional plots viewed from different angles. This capability will be ported to OS/2 and made an integral part of the KBS executive program.

The executive program has a button for each industrial code and facilities to print and browse input and output files generated by the analysis codes. The plotting capability will be added in the future. The user can start any of the codes by clicking on the button for the code. The number of codes running simultaneously is limited only by the amount of memory available on the machine. Multitasking capabilities of OS/2 are exploited to perform printing, browsing and analysis functions in parallel.

The three codes demonstrated included Spiral Groove Gas Seals (SPIRAL), Gas Cylindrical Seals (GCYL) and Incompressible Seals (ICYL). SPIRAL and GCYL interface has the standard PM interface while ICYL interface was designed using ToolBook.

SPIRAL and GCYL have pull down menus to access input screens. The names of the menu items are the same in both programs. Program options area chosen using radio buttons and check boxes instead of entering numbers. Entry fields for numeric information have built in checks to ensure that the numbers are within acceptable ranges. Context sensitive help is provided for all input. When the input is complete, the user may start the analysis by clicking on the analyze button in the menu bar. The output is stored in a use specified output file. The user may also elect to save the input data is a file for future use.

Interactive graphics have been used to ease the input process. In SPIRAL, the user can get immediate feedback during analysis grid specification by clicking on the DRAW button. The program will display the grid as currently specified. In GCYL, the layout of features such as Rayleigh Steps, Recesses, fluid sources, etc. on seal pads is done interactively. The components are laid out on the specified grid using the mouse. This provides a visual representation of the seal pad and avoids having to input a large volume of numerical data. This capability will be extended to include variable grid specification.

ICYL interface shows an alternative approach. The menu items are laid out on the screen in the traditional manner. Submenus appear cascaded on the same screen when a menu item is selected by clicking on it. The concept is similar to the pulldown menus of the standard PM interface. Direct access is provided to any input screen from any other input screen. Menu item names are the same as in SPIRAL and GCYL. Program options and numeric data input procedures are the same as in SPIRAL and GCYL. Arrays of numbers are input using scrollable entry fields. This capability will be added to SPIRAL and GCYL.
SPIRAL-GROOVE CODE

Dr. Jed Walowit

Stator for Spiral Grooved Face Seal
Spiral Grooved Gas Seal Computer Codes

- Shaft seals and face seals
- Compressible flow
- Finite eccentricity and misalignment
- Four degrees of freedom for shaft seals (three for face seals)
- Frequency dependent dynamic coefficients
- Arbitrary end pressures
- Predicts load, flow, power loss, dynamic coefficients, shaft displacements and minimum film thickness

ICYL (Incompressible Cylindrical Seals) Code

Dr. Antonio Artiles

MTI
PROGRAM ICYL

CAPABILITIES

- 2-D incompressible isoviscous flow in cylindrical geometry
- Rotation of both rotor and housing
- Roughness of both rotor and housing
- Arbitrary film thickness distribution, including:
  - Steps, pockets, tapers
  - Preloaded arcs
- Rotor position and velocity is described by four degrees of freedom (translational and rotational)
- External forces and moments may be prescribed to find rotor position
- Pocket pressures or orifice size are prescribed
- Laminar or turbulent flow
- Cavitation
- Inertia pressure drop at inlets to fluid film
  - From ends of seal
  - From pressurized pockets
Computer Code GCYL
(Gas Cylindrical Code)

Wilbur Shapiro

MTI
**Computer Code GCYL**

The general capabilities of the GCYL include:

- Varying geometries
- Variable or constant grid (30 x 74)
- Shaft eccentricity and misalignment
- Specified boundary pressures and periodic boundary conditions
- Symmetry in axial direction
- Determining load (function of displacement) or seal position to satisfy given load
- Choice of English or SI units.

The output of the program include:

- Clearance distribution
- Pressure distribution
- Leakage at specified flow paths
- Load and load angle
- Righting moments
- Viscous dissipation
- Cross-coupled, frequency-dependent, stiffness and damping coefficients
- Plotting routines (pressure and clearance distribution)
Lewis Research Center has a strong history supporting seals, and seal
dynamics, research design, design, fabrication and test. Some efforts include
spiral groove, self energized configurations, Rayleigh lift pads, conical,
labyrinth, bore, honeycomb, face, tip, support for two-phase work, stepped
configurations for SSME, near critical expansion, visualization, dynamic
analyses, and materials.

Some researchers on these topics include: Larry Ludwig and Bob Johnson's
various seal designs and application; John Zuk's cavity analysis and seals
codes; Tom Strom's face and spiral groove seals; Dave Fleming's various
configurations including conical seals and dynamic stability predictions; Bob
Bill's abradable and ceramic configurations; Bill Hughes' two-phase flow
analyses; Hal Sliney's materials work; Gordon Alen, Bill Loomis, El Dirusso
and Bill Hady's test data comparisons and analyses; Isaac Etsion's analyses
of a variety of configurations; Allen Lubeck's face and shaft seal analyses;
Hendricks' shuttle seal and multiple aperture configuration research; Glen
McDonald and Hendricks' ceramic shroud seal; and the Texas A&M program to cite
some of LeRC's efforts.

Some current efforts include: Bruce Steinetz and Paul Sirocky's self sealing
linear segmented ceramic configurations; George Bobula, Bob Bill
and Hendricks' T700 brush seal engine test; flow and duration characteristics of
brush seals and other configurations with Margaret Proctor and Julie
Schlumberger; cryogenic hydrogen brush seal tests at Rocketdyne with Joe
Scharrer; Teledyne brush seal tester with USAF; and support for contracts and
grants.
Primary ring assembly
- Antirotation lug
- Spring
- Carrier
- Secondary ring
- Sealed air $P_1$
- Resilient piloting ring
- Thermal shield

VIEW A-A
FEED GROOVE

VIEW B-B
0.0025 CM
0.001 IN.

DETAINER RING
CARSON RING
SELF-ACTING PAD LOCATION.

VIEW B-8

0.0025 CM
0.001 IN.
LABYRINTH SEAL

SUMP PRESSURE $P_0$

COMPRESSOR PRESSURE $P_1$

LABYRINTH SEAL
Critical mass fluxes of oxygen, nitrogen, argon, and methane computed by isentropic equilibrium expansion using corresponding-states principles.
Schematic of three-step flow path and pressure tap locations. (Linear dimensions are in centimeters (inches).)
Overview of pressure taps and geometry for straight cylindrical seal. (Linear-dimensions are in centimeters (inches).)
Overview of pressure taps and geometry for three-step labyrinth seal. (Linear dimensions are in centimeters (inches), and surface finish on all machined surfaces is 32.)
Reduced mass flux of fluid nitrogen through three-step labyrinth seal in concentric position, as function of reduced inlet stagnation pressure.

Reduced mass flux of fluid nitrogen through three-step cylindrical seal in fully eccentric position, as function of reduced inlet stagnation pressure.
Reduced mass flux of fluid hydrogen through three-step labyrinth seal in concentric position, as function of reduced inlet stagnation pressure.

(a) Fluid hydrogen with an anomaly.
(b) Fluid hydrogen with backpressure control.

Nominal reduced inlet stagnation temperature, $T_{R,0}$
- 0.79
- 0.90
- 1.00
- 1.10
- 1.40

Reduced inlet stagnation temperature, $T_{R,0}$
- 0.35
- 0.30
- 0.25
- 0.20
- 0.15
- 0.10
- 0.05
- 0.00

Tailed symbols denote scatter as effect of backpressure variation.
PRESSURE DROP ACROSS BRUSH VERSUS MASS FLOW RATE
(IN LIQUID NITROGEN)

MASS FLOW RATE, lbm/s/in-circ.

0 0.005 0.01 0.015 0.02 0.025

0 10 20 30 40 50 60 70

PRESSURE DROP, PSI

■ 0 RPM
△ 400 RPM

ORIGINAL PAGE IS OF POOR QUALITY
COMPARISON OF DATA AT 0 AND 400 RPM WITHOUT FLOW STRAIGHTENER
The basic approach for analyzing hydrostatic bearing flows at MSFC is three pronged. First, the Hydrostatic Bearing Team has responsibility for assessing and evaluating flow codes, evaluating friction, ignition, and galling effects, evaluating wear, and performing tests. Secondly, the Office of Aerospace and Exploration Technology Turbomachinery Seals Tasks consist of tests and analyses. Thirdly, MSFC in-house analyses utilize one-dimensional bulk-flow codes; computational fluid dynamics (CFD) analysis is used to enhance understanding of bearing flow physics or to perform parametric analyses that are outside the bulk-flow data base. As long as the bulk-flow codes are accurate enough for most needs, they will be utilized accordingly and will be supported by CFD analysis on an as-needed basis.

Overview

- Hydrostatic Bearing Team formed 02/16/90
  - Assess and validate flow codes
  - Evaluate friction and ignition effects
  - Evaluate wear and galling effects
  - Verify design by HPOTP pump-end test (Rocketdyne IRAD)
    - TTB in October / November timeframe

- OAET Turbomachinery Seals Tasks
  - Three tasks in place -- E3b, E4e, and LSVT13
  - One task pending -- LSVT8 (NRA)

- In-house CFD analyses
  - Baseline damping seal
  - Code validation
  - Rotordynamic coefficients
  - Baseline hydrostatic seal
  - Flow cavity parameters
  - Flow visualization
MSFC Turbomachinery
Working Group Summary

<table>
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<tr>
<th>Work Element Title</th>
<th>Prior</th>
<th>90</th>
<th>91</th>
<th>92</th>
<th>93</th>
<th>94</th>
<th>Product</th>
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<td>E3e. Damping Seal Rotor Support</td>
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<td>Design criteria for rolling element/damping seal assemblies for high side loads; Results led to current TTB validation for HPOTP</td>
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<td>E4e. Damping Seals for Turbomachinery</td>
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<td>Test verification of dynamic properties of damping seals; Wyle tester; measures rotodynamic coefficient density; Test in progress</td>
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<td>LSVT8. Verification of Damping Seal Modeling Techniques</td>
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<td>Hydrostatic bearing data base for code validation; NRA contractor TBD</td>
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<td>LSVT13. Experimental Verification of Rotordynamic Analysis</td>
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<td>Hydrostatic damping bearing data for rotodynamic coefficients; internally and externally fed HPOTP turbine end configuration</td>
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Flight Configuration HPOTP
Phase II
Preburner Pump Bearing Package
HPOTP Pump End Retrofit
Annular Hydrostatic Bearing
Axially Fed

Combined Hydrostatic Bearing/Damping Seal

Experimental Verification of Rotordynamic Analysis
MSFC Program Status

- Complementary Damping bearing development initiated in October
  - Verifies rotordynamic coefficient calculations for hydrostatic bearings

- Test four hydrostatic bearings in modified long life tester
  - HCFC test fluid
    - Two bearings internally fed through the shaft
      - Conventional and damping designs TBD
    - Two bearings internally fed through the stator
      - Conventional and damping designs TBD

- Extracts all rotordynamic coefficients
  - Measures leakage and frictional torque

- Conceptually designs new HPOTP turbine end package
  - Includes lowest whirl ratio bearing tested
  - Provides manufacturing estimate
Potential HPOTP Turbine End
Hybrid Bearing Retrofit

Internally Fed Hydrostatic Bearing/Damping Seal

6 Rotor Feed
Holes Ø 0.219 in DIA. EACH

Shaft Spacer Holes
Angled 50° COUNTER TO ROTATION - DESWIRLS SEAL INLET FLOW

Internally Fed Damping Bearing

Leakage

Leakage

Leakage

Leakage

Inlet Flow

Smooth Land

Axial Gates

Rough Area

OLYP 880518

- George Von Pragenau, NASA-MSFC, Design Concept
Potential HPOTP Turbine End
Hybrid Bearing Retrofit

In-house CFD Analysis

- 3-D analysis; 60° slice of bearing
- Single-phase incompressible Navier-Stokes analysis; constant $\gamma H^2$
- Rotational Reynolds number based on annulus width $\sim 4.8 \times 10^4$
- Multi-block solution in progress with FDNS3D code
  - 3-dimensional pressure-based finite-difference Navier-Stokes solver
  - PISO algorithm with modified Stone’s solver
  - Convection term differencing
    - Central
    - $3^{rd}$-order upwind
    - $2^{nd}$-order upwind
- K-ε turbulence model
  - Two high-Reynolds-number models
  - Four low-Reynolds-number models
In-House CFD Analysis

Configuration
Baseline Hydrostatic Seal

Baseline Damping Seal

In-House CFD Analysis

Typical Grid
Summary

- Hydrostatic Bearing Team meeting regularly with Rocketdyne design organization
  - TTB validation for HPOTP set for October/November 1991 timeframe

- OAET tasks defined
  - Data bases for determining rotordynamic coefficients and flow physics are evolving

- Bulk-flow computer design codes are in place and being extended
  - CFD being applied to support bulk-flow code development for assessing secondary flows in damping seal pockets

- In-house analysis initiated to assess generic flows related to hydrostatic bearings
The Army Propulsion Directorate is primarily concerned with small engine technology, where sealing performance is most critical. Tip leakage and secondary flow losses have a much greater performance impact on small engine aero-components than on large engines. A brief survey and critique of presently employed sealing concepts is presented. Some recent new research thrusts that show promise for substantial improvement are discussed. An especially promising approach for small engine applications is brush seals. Brush seal concepts are being considered for outer air seal and secondary airflow system seal locations.

PRIMARY FUNCTIONS OF ENGINE SEALING

- ISOLATE BEARING COMPARTMENTS FROM ENGINE E
- REDUCE EFFICIENCY LOSSES CAUSED BY LEAKAGE
- CONTROL SECONDARY AIRFLOW SYSTEM
  - COOLING
  - PURGE
  - BUFFER
  - PRESSURE BALANCE
SHAFT SEALS

- ISOLATE BEARING CAVITIES FROM ENGINE ENVIRONMENT
- PREVENT LOSS OF LUBRICANT
(a) SINGLE LABYRINTH, EARLY ENGINES.

(b) MULTIPLE LABYRINTH FOR HIGH-TEMPERATURE HIGH-PRESSURE TURBINE COOLING GAS.

(c) CONVENTIONAL FACE SEAL.

(d) CONVENTIONAL FACE SEAL WITH LABYRINTH SEAL FOR HIGH-TEMPERATURE, HIGH-PRESSURE COOLING GAS.
SUMMARY - SHAFT SEALS

- MUST FUNCTION AS SEALS - NOT FLOW RESTRICTORS
- RUB CAPABILITY AND TEMPERATURE CAPABILITY OF MATERIALS ARE TECHNICAL BARRIERS
- NEW CONCEPTS BEING EXAMINED
  - "BRUSH" SEALS
  - FLUIDIC SEALS

INNER AIR SEALS CONTROL SECONDARY AIRFLOW SYSTEM

- COOLING AIR TO HOT SECTION COMPONENTS
- DISK CAVITY PURGE AIR
- PRESSURE BALANCE SYSTEM
- SHAFT SEAL BUFFER AIR
MAINTAINING MINIMUM CLEARANCE AND KNIFE-EDGE GEOMETRY ARE CRITICAL TO LABYRINTH SEAL PERFORMANCE

SUMMARY - IAS

• CLEARANCE PREDICTION AND CONTROL ARE CRITICAL TO RELIABLE FUNCTION OF IAS SYSTEM
• EXISTING FLOW MODELS ARE CRUDE, SEMI-EMPERICAL
• MATERIAL REQUIREMENTS SIMILAR TO THOSE FOR OAS
• STRUCTURAL STABILITY CONCERNS AND GUIDELINES EXIST
  - AEROELASTIC INSTABILITIES
  - THERMOELASTIC INSTABILITIES
OUTER AIR SEALS REDUCE BLADE TIP LOSSES

- OVER THE TIP LEAKAGE
- INDUCED SECONDARY FLOW LOSSES

Typical compressor efficiency penalty as a function of blade clearance-to-span ratio

Effect of rotor tip clearance on performance for various turbines
OAS MATERIAL REQUIREMENTS

- ABRADABILITY
- EROSION RESISTANCE
- CAPACITY TO SURVIVE IN ENGINE ENVIRONMENT

SUMMARY - OAS

- RETENTION OF MINIMUM TIP CLEARANCES IS CRITICAL TO THE EFFICIENCY OF AERO-COMPONENTS

- OAS SYSTEM MUST MEET CONFLICTING DEMANDS OF:
  - ABRADABILITY
  - EROSION RESISTANCE
  - DURABILITY

- OAS TECHNOLOGY HAS FOCUSED HEAVILY ON MATERIAL APPROACHES
The Turbomachinery Laboratory at Texas A&M University has been providing experimental data and computational codes for the design seals for many years. Dr. Dara Childs began the program with the development of a Halon based seal test rig. This facility provided information about the effective stiffness and damping in whirling seals. The Halon effectively simulated cryogenic fluids. Dr. Childs then developed another test facility (using air as the working fluid) where the stiffness and damping matrices can be determined. This data has been used to develop bulk flow models of the seal's effect upon rotating machinery. In conjunction with Dr. Child's research, Dr. Luis San Andres has developed a bulk flow model for calculation of performance and rotordynamic coefficients of annular pressure seals of arbitrary non-uniform clearance for barotropic fluids such as LH2, LOX, LN2, and CH4. This program is very efficient (fast) and converges for very large eccentricities. Dr. Childs is now working on a bulk flow analysis of the effects of the impeller-shroud interaction upon the stability of pumps.

Dr. Gerald Morrison designed and used this data along with data from other researchers to develop an empirical leakage prediction code for NASA Marshall. He is presently studying, in detail, the flow field inside labyrinth and annular seals. Dr. Morrison is using an advanced 3-D laser Doppler anemometer system to measure the mean velocity and entire Reynolds stress tensor distribution throughout the seals. Concentric and statically eccentric seals have been studied. He is presently studying whirling seals. The data obtained are providing valuable information about the flow phenomena occurring inside the seals, as well as a data base for comparison with numerical predictions and for turbulence model development.

Dr. David Rhode has developed a finite difference computer code for solving the Reynolds averaged Navier Stokes equations inside labyrinth seals. He is currently evaluating a multi-scale k-e turbulence model. Using his computer code, Dr. Rhode designed and patented a new seal geometry. Dr. Rhode is also developing a large scale, 2-D seal flow visualization facility.
HIGH REYNOLDS NUMBER
SEAL TEST SECTION

High-Reynolds-number seal test section

\[
- \begin{pmatrix} F_x \\ F_y \end{pmatrix} = \begin{bmatrix} K & k \\ -k & K \end{bmatrix} \begin{pmatrix} X \\ Y \end{pmatrix} + \begin{bmatrix} C & c \\ -c & C \end{bmatrix} \begin{pmatrix} \dot{X} \\ \dot{Y} \end{pmatrix} + M \begin{pmatrix} \ddot{X} \\ \ddot{Y} \end{pmatrix}
\]
$C_{ef}$ versus $\Delta P$ for sawtooth stators 1 through 4 ($h/C_r = 4.8$), a smooth stator, and the optimum-damping round-hole pattern stator.

$K_{ef}$ versus $\Delta P$ for sawtooth stators 1 through 4 ($h/C_r = 4.8$), a smooth stator, and the optimum-damping round-hole pattern stator.
Test apparatus.

End view of the test-section showing stator instrumentation.
Research Objective

To obtain a better understanding of the flow field inside annular and labyrinth seals.

This information is important for:

1. Gaining insight into the turbulent flow fields, how they change with operating conditions, and how they effect leakage and stability.

2. Providing detailed flow field measurements for the purpose of comparing to and helping refine computational predictions of the flow field.

3-D LASER DOPPLER VELOCIMETER

Three Colors: Green, Blue, Violet
Three Bragg Cells: 40 MHz
8.5X and 3.75X Beam Expanders
450 mm Lenses
1 X 1 X 4 Mil Measurement Volume

Simultaneously Measures:
Mean Velocity Vector
U, V, W, with Flow Reversals
Entire Reynolds Stress Tensor
u'u', v'v', w'w', u'v', u'w', v'w'

3-D Traverse System
Rotary Encoding System for Periodic Flow Mapping
Integrated Data Acquisition, Analysis, and Traverse Control
The 3-D LDA system is uniquely qualified for this study due to:

1. The small size of the probe volume (0.001" X 0.001" X 0.004").

2. The non-invasive nature of the measurement device.

3. The ability to measure the mean velocity and the entire Reynolds stress tensor.

4. Ensemble averaging capability for use on whirling rotors.

5. The ability to measure flow reversals.
Mean Velocity Vector Fields, $\theta = 0^\circ$ and $180^\circ$. 

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Mean Azimuthal Velocity Contours, \( \frac{U_{\varphi}}{W_{sh}}, \vartheta = 0^\circ \) and 180°.
Turbulent Kinetic Energy Contours, $k / U^2$, $\theta = 0^\circ$ and $180^\circ$.
Reynolds Stress Contours, $u'_x u'_y/U^2$, $\theta = 0^\circ$ and $180^\circ$. 

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**Measurements To Date:**

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**Measurements To Be Made This Year**

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Flat plate tester assembly.
A typical friction-factor pattern.

Frequency spectra for test number 7.

(b=1.57, d=3.05 and H=0.25mm)
MEASURED LABYRINTH LEAKAGE REDUCTION FROM CFD DESIGN (Rhode, et al.)

Measured leakage rate characteristic for the impeller inlet seal of the space shuttle fuel pump comparing the new CFD design with the current design.

NEW, ADJUSTABLE-GEOMETRY LABYRINTH RIG:
LEAKAGE & FLOW VISUALIZATION

Sketch of the quick-change geometry labyrinth seal water test rig
Correlate Large-Scale With Full-Scale Design Results

A. Design and Construct Rig (Variable c and Rotor Position)
B. Fabricate Several Advanced Land and Knifed Surfaces
C. Obtain Scale Relationships for Limiting Cases:
   1. With and Without Rub Grooves
   2. Standard and Sharp Corners
   3. Standard and Advanced Knife Tip Shapes

Advanced Geometry Effects on \( \dot{m} \) and \( \dot{m} \)-Variation

A. Standard Corners (\( R=0.003 \) in)
   Obtain design data, including \( \dot{m} \) Variation
   with a change of: (a) clearance and (b) relative
   rotor position for:

Record:
\( \dot{m} \)
VCR Movies
Photographs
F.D. Solution
Five-Hole Pitot

1. Baseline geometry
2. Various annular grooves
   (stator and stator + rotor)
3. Various step heights s
4. Various knife thickness \( t_2 \)
5. Various knife angles

B. Sharper Corners (\( R=0.0015 \) in)
   (Same as A)

Record:
\( \dot{m} \)
VCR Movies
Photographs
F.D. Solution
Five-Hole Pitot

1. Baseline geometry
2. Various annular grooves
   (stator and stator + rotor)
3. Various step heights s
4. Various knife thickness \( t_2 \)
5. Various knife angles
Research Objectives on Annular Seals for 1991–92 at Texas A&M University

P.I. Dr. L. San Andres

- Analysis of thermal effects on the performance and dynamic force response of high pressure annular seals handling cryogenic liquids. Adiabatic and Constant Surface Temps. Important for LOX due to higher viscosity and lower specific heat than LH2.
- Development of a unified theory for liquid, gas, and two-phase flow annular seals.
- Calculation of dynamic force and moment coeffs. for displacements and angular motions.
- Analysis of high pressure OIL SEAL RINGS for compressors: single and multi-land seals. Calculation of lock-up forces and face friction factors.
- Study of Damper Seals and Hydrostatic Bearings as support elements in cryogenic turbopumps.
- Improved Seal Entrance Factors by CFM simulations.
Analysis of Turbulent Flow
Annular Pressure Seals

Dr. L. SanAndres, Texas A&M University

- 2-D Bulk Flow Model for barotropic liquids
  Surface roughness: Moody's friction factor

- Arbitrary Axial Clearance Distribution:
  uniform, taper, step, wavy via spline, and stator wear profile

- Arbitrary Static Eccentricity Ratio, 0 ≤ e/c ≤ 1

- Realistic liquid properties for LH2, LN2, LOX, CH4

**Program calculates:**
- Leakage, Fluid Film Forces, Torque, and Rotordynamic Coefficients: Kij, Cij, Mij

- Solution scheme based on efficient CFM methods and improved approximate guesses for fast convergence.

- Program can be used for flooded or grooved journal bearings.

- Laminar and/or Turbulent Seals and Bearings
SEALS RESEARCH AT THE UNIVERSITY OF AKRON

M.J. Braun
University of Akron
Akron, Ohio

FLOW AND PRESSURE STUDIES ON BRUSH SEALS

The flow visualization effort aims at determining the flow regimes and the sealing mechanisms that are at work in the brush seal, regardless of whether the seal is linear or circular. A vision system coupled with an image processing system is used to analyze the nature of the flow.

Pressure measurements were made to study the effect of the number of brushes as well as the brush structure on the seal effectiveness.

NUMERICAL SIMULATION OF FLOW IN A HYDROSTATIC POCKET

This portion of the presentation will detail a numerical approach to flow simulation in one hydrostatic pocket. The effects of the jet, shaft velocity and clearance on the flow patterns and pressure generation will be described.
The Lubrication Branch has two active programs that are developing gas turbine engine mainshaft air/oil seals. Both of these programs, one of which is with General Electric Aircraft Engines and the other with Pratt & Whitney Aircraft, are addressing counter-rotating intershaft applications which involve very high rubbing velocities. My presentation will briefly address the objectives and requirements of both of these efforts.

**HIGH SPEED AIR/OIL SEAL DEVELOPMENT**

**CONTRACTOR:** UNITED TECHNOLOGIES (PRATT & WHITNEY)

**CONTRACT NUMBER:** F33615-88-C-2822

**START / DURATION:** SEP 88 / 45 MONTHS

**OBJECTIVE:** CONDUCT A COMBINED ANALYTICAL AND EXPERIMENTAL PROGRAM TO DEVELOP AND ENDURANCE TEST COUNTER-ROTATING INTERSHAFT SEALS FOR RELIABLE OPERATION UP TO 1200 FT/SEC PITCH LINE VELOCITY

**GOALS:**
* 4000 HOUR SEAL LIFE
* 1200 FT/SEC PITCH LINE VELOCITY
* 50 LB/HR MAX AIR LEAKAGE @ 50 PSID & 750 F
* 500 LB/HR MAX AIR LEAKAGE AFTER PRIMARY SEAL MALFUNCTION
* MISALIGNMENT TOLERANCE
HIGH SPEED AIR/OIL SEAL DEVELOPMENT

APPROACH:

* DEFINE THE SEAL APPLICATION AND ESTABLISH REQUIREMENTS
* SELECT TWO SEAL CANDIDATES THROUGH AN ANALYTICAL ASSESSMENT OF PERFORMANCE POTENTIAL FROM A MINIMUM OF FIVE SEAL DESIGNS
* FABRICATE AND EVALUATE THE PERFORMANCE CAPABILITIES OF BOTH SEAL DESIGNS
* SELECT THE BETTER OF THE TWO SEAL DESIGNS FOR ENDURANCE TESTING
* ENDURANCE TEST THIS SEAL FOR AN ADDITIONAL 80 HOURS
* EVALUATE TEST RESULTS TO IDENTIFY IMPROVEMENTS NEEDED AND POSSIBILITY OF TRANSITION INTO THE JTDE PROGRAM

HIGH SPEED AIR/OIL SEAL DEVELOPMENT

STATUS:

* SEAL DESIGN REQUIREMENTS WERE BASED ON AN ADVANCED FIGHTER-TYPE AIRCRAFT WITH A 60% INCREASE IN THE THRUST-TO-WEIGHT CAPABILITY OF CURRENT STATE-OF-THE-ART SYSTEMS
* TWO SEAL DESIGNS, A CONTROLLED GAP SEAL CONCEPT AND A SEGMENTED HYDRODYNAMIC CIRCUMFERENTIAL SEAL, WERE SELECTED FROM 25 CANDIDATES
* FABRICATION OF THE CONTROLLED GAP SEAL DESIGN IS IN PROGRESS
* FABRICATION OF THE SEGMENTED SEAL IS COMPLETE
* SHAKE-DOWN TESTING OF THE EG&G SEALOL TEST RIG HAS BEEN COMPLETED
COUNTER-ROTATING INTERSHAFT SEAL DEVELOPMENT

CONTRACTOR: G.E. AIRCRAFT ENGINES

CONTRACTOR NUMBER: F33615-90-C-2000

START/DURATION: APR 90/29 MONTHS

OBJECTIVE: DEVELOP AN ADVANCED DESIGN INTERSHAFT SEAL FOR PHASE II IHPTET ENGINE CONFIGURATIONS THAT UTILIZE COUNTER-ROTATING TWIN SPOOLS

GOALS:
- 900 FT/SEC PITCH LINE VELOCITY
- Δ P UP TO 50 PSID
- 900°F SEAL AIR INLET TEMPERATURE

COUNTER-ROTATING INTERSHAFT SEAL DEVELOPMENT

APPROACH:

- ADVANCED DESIGN BASED ON AN AIR BEARING-SUPPORTED CONTINUOUS RING CONFIGURATION

- THREE MATERIAL APPROACHES FOR CRITICAL SEALING INTERFACE:
  
  (1) CONCURRENTLY DEVELOPED ADVANCED HIGH STRENGTH CARBON (> 100KSI PER LB/IN^3)
  
  (2) STATE-OF-THE-ART METALLIC MATERIALS
  
  (3) INLAID ARRANGEMENTS OF CARBON, METAL, AND/OR CERAMICS
COUNTER-ROTATING INTERSHAFT SEAL DEVELOPMENT

APPROACH (CON'T):

- CONDUCT PHYSICAL AND TRIBOLOGICAL CHARACTERIZATION TESTS TO SELECT BEST MATERIAL APPROACH

- FOR SELECTED APPROACH, DESIGN AND FABRICATE FULL-SIZE PROTOTYPE SEAL

- CONDUCT SEAL TESTS IN TWO PHASES:
  1. PERFORMANCE MAPPING
  2. ENDURANCE FOR UP TO 200 HRS

- BASED ON TESTS, DESIGN SEAL FOR THE PH II DEMONSTRATOR ENGINE

COUNTER-ROTATING INTERSHAFT SEAL DEVELOPMENT

STATUS:

- MATERIAL CHARACTERIZATION TESTS UNDERWAY

- PROMISING RESULTS OBTAINED WITH ADVANCED CARBON MATERIALS
  1. HIGH STRENGTH
  2. GOOD OXIDATIVE RESISTANCE
Brush Seals

Brush seals are being evaluated for potential near and far term gas turbine engine applications. Development is in the form of rig component testing, and engine testing. Allison has tested an engine with 20 individual brush seal positions. These seals were located throughout the engine. The emphasis of the current work is on obtaining long term performance data for brush seals. Very little of this data is available.

Film Riding Face Seals

Allison is presently developing film riding face seal technology to support future gas turbine engine applications. Face seal with an approximate 7" diameter has been successfully tested to 1000°F, 100 psid, and 650 ft/sec. Seal leakage remained below 1 scfm throughout the duration of the test. A model for the compressible gas film has been developed which separates the primary seal rings during operation. This model is based on the traditional Reynolds's approach which is customarily applied to lubrication type problems. Because of the difficulty of experimentally verifying the program predictions, a commercial Navier-Stokes code was used in parallel. By comparing predictions for similar cases, it is expected that the limitations of the Reynolds's model can be assessed as it applies to this particular seal.
About four years ago, work was completed on a 36 inch diameter gas to gas carbon ring seal used to buffer low pressure turbine air at the rim of the forward outer flowpath on the GE36 unducted fan (UDF) engine. This highly successful program demonstrated the ability to provide long life at a gas leakage rate equivalent to a .0018 inch clearance, or, conservatively estimating, a 20 to 1 reduction over a labyrinth, and an SFC reduction of at least 2.5%. Operating conditions were 1600 rpm (240 fps), 60 psid and 650 deg F. The seal design was based on the use of self-acting gas hydrostatic bearings.

At about the same time, we were working an AF/Navy contract for development of a long life counter-rotating intershaft air-oil seal of approximately 7.6 inch diameter for operation at 800 fps, 800 deg F and 50 psid. Although we were successful in meeting most program goals with a split ring seal of the axial bushing type, the seal with the greatest payoff in life and air leakage rates, bearing many features common with the GE36 seal, could not be successfully tested because of the structural weakness of the primary seal ring carbon material. This was a split ring seal using a hybrid combination of orifice compensated hydrostatic and shrouded hydrodynamic gas bearings. We are presently working an AF contract to develop this design in conjunction with high strength carbon materials being developed by Pure Carbon Co.

In the area of engine secondary gas flow path sealing for performance improvement, we are currently working with carbon and all metal face seals. Nine inch diameter hydrodynamic carbon face seals at 450 fps, 140 psid and 950 degrees F, have demonstrated long life at a flow reduction of approximately 96% (7.5 scfm) compared to a "best" labyrinth. A fifteen inch diameter all metal "aspirating" face seal, using self-acting hydrostatic bearings, has been successfully tested to 700 fps, 100 psid and 1000 deg F, demonstrating long life at a flow reduction of 86% compared to a "best" labyrinth. This seal will be developed through 1400 deg F, 900 fps and 350 psid. The seal "aspirates" closed at about idle speed pressure during engine start and reopens at engine shutdown.

A hydraulic thrust balance "seal", currently using orifice compensated hydrostatics, is under development. We have demonstrated long life operation for most large engine low pressure rotor applications (170 fps) with 30000 pounds control at 7 inches diameter and less than 3 gpm flow rate using Type 2 (23699) engine oil. We are now working high pressure rotor systems, to 450 fps. This program has two significant payoffs: 1) as much as 50 to 1 increase in thrust bearing fatigue life, 2) as much as 3% reduction in SFC and significant multipliers in turbine bucket life.
CROSS SECTION OF A 7 INCH HYDRAULIC THRUST CONTROL.

An axial force is transmitted to the engine rotor by applying oil under pressure to the hydraulic area of the thrust control mechanism.
ASPIRATING GAS BEARING FACE SEAL

CONTINUOUS RING SEAL DESIGN
LARGE DIAMETER HYDROSTATIC SEAL

1 Primary Seal Elements
2 Outer Race, Rotating
3 Fwd Race, Stationary
4 Aft Race, Stationary
5 Axial Springs
6 Piston Ring, Stationary
7 Radial Bushings
8 Piston Ring, Rotating
9 Radial Splines, Clamp-Race
10 Carrier, Stationary Piston Ring
11 Clamp Ring
12 Clamp Spring
13 Adapter, Seal to Frame
14 Flange, Engine Frame
15 Radial Splines, Fwd-Aft Races
16 Retainer Lugs, Carbon Segment
A brief introduction is made describing EG&G Fluid Components Technology Group and FCT Group R&D capabilities.

Seal related development activities, including modeling, analysis and performance testing are described for several current seal related projects. These include noncontacting, high speed, high pressure gas sealing systems for turbomachinery, brush seals for gas path sealing in gas turbines and tribological material evaluation for wear surfaces in sealing systems.

GROUP R & D

Staffed and equipped for:

- Material Characterization and Analysis
- Tribological Characteristics, Testing and Evaluation
- High Temperature, High Pressure and High Speed Testing of Seals and Sealing Systems
- Mechanical Design of Products and Specialized Testing Equipment
- Modeling of Systems and Components, Design Analysis and Performance Parameters
- Manufacturing Process Development
ADVANCED AEROSPACE
TEST RIG

- Rig Capabilities (Counter Rotating Configuration)
  - Spindle Speed: 24,000 RPM each shaft
  - Power: 43 HP at 17,000 RPM, 32 HP at 26,000 RPM
  - Temperature of Test Chamber: 600°F
  - Test Chamber Pressure: 100 PSI
  - Axial Travel: ±.0625 inch each shaft

- Single Shaft (Brush Seal Testing)
  - Spindle Speed: 24,000 RPM
  - Power: 43 HP at 17,000 RPM, 32 HP at 26,000 RPM
  - Temperature of Test Chamber: 100°F
  - Test Chamber Pressure: 100 PSI
  - Axial Travel: ±.0625 inches
SEAL RELATED DEVELOPMENT ACTIVITIES

BRUSH SEAL DEVELOPMENT PROGRAM
U. S. AIR FORCE PRDA-11
IN-HOUSE BRUSH SEAL TESTING

NON-CONTACTING GAS FILM SEALS
DEVELOPMENT AND TESTING

TRIBOLOGICAL MATERIAL EVALUATION
ANALYSIS AND TRIBOLOGICAL TESTING

Brush Seal Development Program
U. S. Air Force
Integrated High Performance Turbine Engine Technology — IHPTET
PRDA —II
Objective: Develop a comprehensive design methodology for Brush Seals using
1. Application requirements from engine manufacturer
2. Experimental characterization of seal design and tribological pairs

Goals:
1. Substantially lower leakage than laby seals
2. Seal life consistent with man—rated mission requirements
3. Achieve maximum pressure sealing capability with single or multiple stages

Provide:
1. Comprehensive seal design manual
2. Man—rated brush seal for test in an IHPTET engine

Project duration is two and one half years, started July 1990
PACK WIDTH  
BACKING WASHER  

FREE LENGTH  
MIN. TYP  

RETAINING WASHER  
COVER PLATE OR SIDE PLATE  

WIRE DIAMETER - d  
TOTAL NUMBER OF BRISTLES - n  
WIRE DENSITY - \( \rho \)  

**BRUSH SEAL ASSY.**  

FLOW PARAMETER VS DP/P. COMPARISON OF BRUSH SEAL AND LABYRINTH SEAL LEAKAGE  

**LEGEND**  
\( \circ \) = EPD TESTS 5500 RPM  
\( \circ \) = EPD TESTS 9700 RPM  
\( \triangle \) = EPD TESTS 11600 RPM  
\( + \) = EPD EH TESTS 6K-15K RPM  
\( \times \) = TELEDYNE TESTS 30000 RPM  
\( \diamond \) = EPD 6 FIN LABY (.035 GAP)  
\( \triangledown \) = ALLISON 4 FIN (.02 GAP)  

*ORIGINAL PAGE IS OF POOR QUALITY*
Metal Bellows Gas Seal (Patent Pending)
(Buffer Gas Arrangement)
Dynamic Test Arrangement of a Thinline Gas Seal

Energy Dissipation vs. Pressure

Single, data per pod

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<th>SPEED</th>
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15500 RPM
Leakage vs. Pressure

Single, data of POD 1

Leakage Correlation

Single, data of POD 1
Unique Tribology Test Rigs

High Temperature
Oscillating Test Rig

Two unique test rigs have been designed to evaluate the wear and friction of materials for aerospace sealing applications.

The first is a high temperature, oscillating rig used for tribological testing of materials for slide joints, static seals, solenoid valve plungers and similar aerospace applications exhibiting fretting type wear. It was designed and built at Group R&D to accurately record variations in the normal and tangential loads under oscillating conditions at temperatures up to 1400°. The design of this rig uses a programmable motor which can duplicate acceleration and velocity profiles of various oscillating and low speed unidirectional motions. The test samples are enclosed in a furnace which can be purged using various atmospheres. A pneumatic loading system is designed to achieve high contact pressures (500 psi), applied pneumatically.
High Speed-High Temperature Test Rig

The second is a high speed/high temperature rig designed to test materials for main shaft and gas-path seal applications where high surface speeds of up to approximately 400 feet per second at a shaft rotation of 60,000 rpm and light loads up to 2psi are used for testing. This rig has the capability of heating the tribo-pairs to 1200° while maintaining precise alignment at the seal interface. Loading of the seal face is accomplished using gas pressure applied to a bellows assembly such that precise increments of pressure can be selected for test purposes. This test rig is currently being used to investigate appropriate tribo-pairs for brush seal fibers and runners.

FRICTION OF CARBON GRAPHITE AT HIGH TEMPERATURES

![Graph showing friction coefficient vs temperature for different materials](image-url)
WEAR OF CARBON GRAPHITE AT HIGH TEMPERATURE

Temperature (C)

Wear (mgs)

MANF. TEMP. LIMIT FOR A

MANF. TEMP. LIMIT FOR B

CARBON GRAPHITE B

CARBON GRAPHITE A
By definition differential roughness occurs where the surface roughness of one region is distinctly different (more or less) than that of an adjacent region. Differential roughness may occur on a sliding surface when the structure of an area of the surface is modified relative to the original material. Differential roughness may be used to effect a film thickness change so as to cause hydrodynamic or hydrostatic lubrication effects just as if the surfaces were machined. The advantage of differential roughness is that the effective offset of film thickness continues to exist even if there is gross wear. One can effect film thickness changes which are smaller than can be made directly. Furthermore, asperity tip load support is the same over both the high roughness and small roughness regions.

The potential uses for differential roughness are seals, bearings, and pumps. Some examples are presented.
Rocketdyne Seal Technology

- Rocketdyne has a large collection of seal analysis codes
  - Floating ring seals
    - Static equilibrium
    - Dynamic response
  - Face seals
    - Steady state performance
    - Stability analysis
  - Load sharing seals
    - CFD codes for labyrinth and annular seals
    - Bulk-flow analysis for rotordynamic coefficients
Rocketdyne Seal Technology

• Rocketdyne code development is ongoing
  - Barotropic fluid properties
  - Phase change
  - Brush seal performance
  - Large scale roughness
  - Asymmetric boundary conditions

• Rocketdyne has both contract and internal funding for seal testing
  • Contract efforts
    - AFAL LOX/H2 Turbomachinery Technology
      - Extraction of rotordynamic coefficients for knurled damping seal in LH2
      - Dynamic testing of floating ring seal in LH2
      - Dynamic testing of spiral groove face seal in LH2
    - NASP
      - Extraction of rotordynamic coefficients for knurled damping seal in LOX
    - NASA MSFC NRA (with University of Akron)
      - Flow visualization and measurement in knurled damping seal
Rocketdyne Seal Technology

- IR&D funded testing
  - Brush seal performance in LH2
  - Brush seal coatings (with Texas A&M)
  - Flow in large scale roughness (with Texas A&M)
  - Transient load capacity of knurled damping seal in LOX
Annular Hydrostatic Bearing Transient Tester

- Radial Load Applicator
- Fluid Supply
- Journal
- Dual Ball Bearing Pack
- System Centering Support
- Annular Hydrostatic Bearing

HPOTP PUMP END BEARING CONVERSION Plans

- Design annular LOX hydrostatic bearing/seal
  - Tapered, knurled sterling silver damping bearing
- Eliminate ball bearings No. 1 and No. 2
- IR&D to design and fabricate retrofit hardware
- Contract to assemble and test on TTB
- Design substantiated with component tests in Long Life tester
  - K-monel and Inconel 718 journals
  - 60 start transient simulations per bearing in LOX
- Tribometer tests measure friction and wear in GOX
- Ignition data provided by White Sands testing
- Future plans include upgrading tester to measure coefficients
HPOTP PUMP END BEARING CONVERSION

Flight Phase II Configuration

Hydrostatic Bearing Retrofit

- Twelve parts & bolt pattern removed
- Replaced by three parts

EXPERIMENTAL VERIFICATION OF ROTORDYNAMIC ANALYSIS

MSFC Program Status

- Complementary damping bearing development initiated in October
  - Verifies rotordynamic coefficient calculations for hydrostatic bearings

- Tests four hydrostatic bearings in modified long life tester
  - HCFC test fluid

  - Two bearings internally fed through the shaft
    - Conventional and damping designs TBD

  - Two bearings externally fed through the stator
    - Conventional and damping designs TBD

- Extracts all rotordynamic coefficients
  - Measures leakage and frictional torque

- Conceptually designs new HPOTP turbine end package
  - Includes lowest whirl ratio bearing tested
  - Provides manufacturing estimate
HYDROSTATIC BEARING TECHNOLOGY
Review

- **Problem**
  - Present turbopumps limited by rolling element bearing life

- **Goal**
  - Develop reliable, long life, cryogenic fluid-film bearings

- **Benefits**
  - Eliminate rolling element bearing DN limits and life constraints
  - Add flexibility to optimize rotor mechanical arrangement
  - Increase damping for more stable operation
  - Reduce overhauls and maintenance costs

- Rocketdyne committed to multiyear IR&D development program
  - 1990 expenditure of $1.726M
  - 1991 commitment of $2.275M

- NASA and AFAL contracts in place to further advance technology

### HYDROSTATIC BEARING DEVELOPMENT INTEGRATION PLAN

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This presentation package covers the CFD activity at Aerojet related to seals and fluid film bearing. The presentation addresses the following topics:

1. Aerovisc Numeric and Capabilities
2. Recent Seal Application
3. Future Code Development

PRESENTATION PREVIEW

- AEROVISX NUMERICS AND CAPABILITIES

- RECENT SEAL APPLICATION

- FUTURE CODE DEVELOPMENT
AEROVISC Numerics

- **Formulation**
  - Reynolds Stress Averaged Navier-Stokes Equations in Cartesian, Strongly Conservative, Primitive Variable Form
  - k-e and ARS Turbulence Models With Log-Law Wall Functions

- **Discretization**
  - "Flux" Element Based Finite Volume Method
  - General Non-Orthogonal Boundary-Fitted Structured Grid
  - Choice of Advection Schemes
    - Upward Difference (Most Robust, Least Accurate)
    - Mass Weighted Skew (Enhanced Accuracy)
    - Linear Profile Skew (Most Accurate)
  - Second-Order-Accuracy With Physical Advection Correction Term
  - Rhie-Type Pressure Redistribution for Incompressible Flows

- **Algebraic Solver**
  - Choice of Vectorized Gauss-Siedel or Incomplete Cholesky Base Solver
  - Additive Correction Multigrid (Large Grids)
  - Block Correction (High Aspect Ratio Grids)

**RELEVANT CODE CAPABILITIES**

- INCOMPRESSIBLE FLOW
- SUBSONIC, TRANSONIC, AND SUPERSONIC FLOW
- NON-ISOTHERMAL AND ISOTHERMAL FLOW
- LAMINAR, TURBULENT, OR INVISCID FLOW
- CORIOLIS AND CENTRIFUGAL TERMS FOR TURBOMACHINERY APPLICATIONS
- FIXED, MOVING OR ROTATING TURBULENT WALLS
- CONJUGATE HEAT TRANSFER or SPECIFIED WALL TEMPERATURE/FLUX
- VARIABLE FLUID AND SOLID PROPERTIES
- MULTI-COMPONENT FLOW (N ADDITIONAL SCALAR TRANSPORT EQUATIONS)
- MULTIPLE BLOCKED REGIONS
FUTURE CODE DEVELOPMENT

- GRID EMBEDDING/ATTACHING
  - GRID REFINEMENT IN AND NEAR SEALS
  - IMPROVED SOLUTION ACCURACY

- MULTI-LAYER TURBULENCE MODEL

- AUTOMATED PROCEDURE TO PREDICT FLUID SEAL DYNAMIC COEFFICIENTS
Schematic of Fluid Film Bearing Analysis Methodology
An example of CFD application in the seal area is described. This effort consisted of optimizing specific seals in SSME-ATD and resulted in design of a seal tested at Pratt and Whitney which demonstrated 30% lower leakage than standard design at the same clearance. This example shows the potential of identifying seal geometry parameters which reduce internal flow leakages detrimental to component and turbopump efficiencies. This information can then be used to develop an analytical design procedure for controlling leakage in any given application or environment.
COMPUTATIONAL FLUID DYNAMICS

Significant Computing Power Available

Cray X-MP-2/8

P&W

SUN WORKSTATIONS

Numerical Aerodynamic Simulator (NAS)

NASA LeRC

NASA MSFC

APL

Silicon Graphics "IRIS" Workstations

2-D, 3-D Graphics

P&W

IBM SYSTEM

CFD ANALYSES OF TURBOPUMPS

Seven Areas Of Analysis Identified As Important For Successful Turbopump Design

Fuel pump interstage
diffuser/crossover

Turnaround duct/hot gas
manifold interface

Bearings/seals

Inlet scroll

Preburner
temperature
distribution

Oxidizer pump

Turbine rotor/stator

interaction

Turbine temperature
profile redistribution

HPFTP

HPOTP
SSME-ATD FUEL PUMP CAVITY
Minimize Leakage

Overall original BI Seal configuration.

Streamline pattern within the idealized original cavity.

Design dimensions for the BI Seal configurations.

Streamline pattern within the Seal B cavity.

ATD HPOTP INTERNAL FLOW MGMT
Bearing And Turbine Coolant Flows

Roller Bearing Coolant Flow

ORIGINAL PAGE IS OF POOR QUALITY
ATD HPOTP
Seal And Bearing Details

SSME-ATD LOX PUMP CAVITY
Controlled Leakage

Overall original Interpropellant (IP) Seal design.

Recommended design configuration for the IP Seal utilizing the new leakage stability concept.

Streamline pattern within Seal 75 cavity.
CFD Applications in Propulsion

Charles L. Merkle
The Pennsylvania State University
Department of Mechanical Engineering
University Park, PA 16802

An overview of various applications of CFD algorithms to propulsion problems is given. Problems of interest include incompressible, low speed compressible, transonic and supersonic. A common family of algorithms is used for all applications and emphasis is placed on maintaining accuracy and convergence efficiency for all problems. Specific problems include pump hydrodynamics, combustion and mixing simultaneous in rocket engines, viscous nozzle flow, and CFD applications to combustion stability.

CURRENT PROJECTS

• ROCKET COMBUSTOR MODELING
• COMBUSTION INSTABILITY MODELING
• PUMP FLOWFIELDS
• VISCOUS NOZZLE/PLUME FLOWS
• MAXWELL/NAVIER-STOKES ANALYSIS
• AUXILIARY PROPULSION
• LOW SPEED COMPRESSIBLE FLOWS
CFD PROBLEM FORMULATION

\[ \Gamma \frac{\partial \rho}{\partial t} + \frac{\partial E_i}{\partial x_i} = \frac{\partial}{\partial x_i} \left( R_{ik} \frac{\partial \tilde{\phi}_k}{\partial x_j} \right) + H \]

Steady Compressible
Unsteady Incompressible
Viscous Upwind
Inviscid Central
Schematic of small thruster geometry.

Pressure Contours

\[ \text{Min} = 23.4 \text{ kPa} \]
\[ \text{Max} = 304 \text{ kPa} \]
\[ \Delta \text{P} = 20 \text{ kPa} \]
TURBULENT REACTING FLOW

TEMPERATURE CONTOURS

\[ \text{MIN} = 346.9 \, \text{K} \]
\[ \text{MAX} = 3302 \, \text{K} \]
\[ \text{DELTA} = 250 \, \text{K} \]

SPECIES MASS FRACTION \( Y_i \)
Plunging Airfoil
OVERVIEW

The CWRU Seal/Bearing test facility is shown in Figures 1 through 3, and the revised force measuring system is sketched in Figure 4. This facility has recently been retrofitted with a high-pressure high-flow oil system, which was acquired to conduct basic research on hydrostatic journal bearings for NASA Lewis Research Center. Mr. Russell Capaldi is the NASA grant monitor. The high-pressure high-flow water system remains in place to test seals. Also, a new high-flow air system is now installed. Thus, testing to determine static and dynamic properties can now be performed using oil, water or air on this single facility.

We are currently using the oil system to determine rotordynamic properties of a NASA four-pocket hydrostatic journal bearing. The revised dual system force measuring configuration (see Figure 4) is performing with excellent accuracy. That is, the dynamic force measurements are made simultaneously with two independent systems, one with piezoelectric load cells and the other with strain gage load cells. The difference is less than 2% (see Figure 5 through 7) between these two sets of load cell measurements on recent tests with a static eccentricity set close to zero (e= 0.001 inch, C = 0.009 inch) and an orbit radius of 0.0004 inch. Table 1 shows the extracted stiffness, damping and inertia coefficients for the test conditions shown, as extracted from the two independent dynamic force measurements. These coefficients were extracted using a linear-regression least-squares fit of the dynamic force and orbit displacement signals over a frequency range of 250 to 2,400 cpm, without constraining the inertia matrix to be symmetric.
1(a) Conceptual Sketch of Rotor Support Component Test Apparatus

1(b) Assembly Layout of Rotor Support Component Test Apparatus

Figure 1. Test Apparatus
Fig. 2. Floor plan and elevation view of test facility.
Fig. 3. Schematic of data acquisition system.
Figure 4(a) New load measuring and support system with strain gage transducer.

Figure 4(b) New load measuring system, showing complete assembly with water-seal barrier.
TOTAL TIME-AVERAGED SIGNAL (50 cycles)

Figure 5

Single-Peak Amplitude and Phase
Angle of Fourier Fundamental Term:

\[ \omega = 1.000 \text{ RPM} \]
\[ \Omega_{	ext{base}} = 675 \text{ CPM} \]
\[ P_{	ext{base}} = 75 \text{ psi} \]

\[ 20.301 \text{ LDF} \]
\[ -24.109 \text{ DEGREE} \]

TOTAL TIME-AVERAGED SIGNAL (50 cycles)

Figure 6

Single-Peak Amplitude and Phase
Angle of Fourier Fundamental Term:

\[ \omega = 1.000 \text{ RPM} \]
\[ \Omega_{	ext{base}} = 675 \text{ CPM} \]
\[ P_{	ext{base}} = 75 \text{ psi} \]

\[ 20.005 \text{ LDF} \]
\[ -25.736 \text{ DEGREE} \]
FIRST TERM OF FOURIER DECOMPOSITION

\[ \omega_0 = 1,000 \text{ RPM} \]
\[ \omega_{\text{d}} = 675 \text{ CPM} \]
\[ \rho_\text{m} = 75 \text{ psig} \]

\[ \text{TIME (\text{\degree})} \quad \text{PERIOD (\text{DEG})} \]

Dynamic Coefficients Report

Special File Name is: CCF.INP
Total Input FS Data File is: 10
Shaftspeed (RPM) is: 1000
Recess Pressure (PSI) is: 75

Input Data Filename | No. of Cycle | Orbit Frequency (RPM)
-------------------|--------------|---------------------
CC1F.FS            | 1            | 147                 
CC2F.FS            | 1            | 262                 
CC3F.FS            | 1            | 376                 
CC4F.FS            | 1            | 569                 
CC5F.FS            | 1            | 677                 
CC6F.FS            | 1            | 785                 
CC7F.FS            | 1            | 871                 
CC8F.FS            | 1            | 1041                
CC9F.FS            | 1            | 1139                
CC10F.FS           | 1            | 1247                

Potroponic Coefficients:

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Table 1(a)
DYNAMIC COEFFICIENTS REPORT

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TOTAL INPUT FS DATA FILE IS: 10
SHAFTSPEED (RPM) IS: 2000
RECESS PRESSURE (PSI) IS: 75

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TRANSFERENCE DYNAMIC COEFFICIENTS:

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Table 1(b)
MIT reports on a seal test program that was conducted in a dynamic rig. Four different 1-gland labyrinths were tested. The eccentricity, spin rate, whirl rate and flow were varied independently.
Key to the successful development of the single-stage-to-orbit National Aerospace Plane (NASP) is the successful development of combined cycle ramjet/scramjet engines that can propel the vehicle to 17,000 mph to reach low earth orbit. To achieve engine performance over this speed range, movable engine panels are used to tailor engine flow that require low-leakage high-temperature seals around their perimeter. NASA Lewis is developing a family of new high temperature seals to form effective barriers against leakage of extremely hot (>2000 °F), high pressure (up to 100 psi) flow path gases containing hydrogen and oxygen. Preventing backside leakage of these explosive gas mixtures is paramount in preventing the potential loss of the engines or the entire vehicle.

Described in the subsequent pages of this report are seal technology development accomplishments in the three main areas of concept development, test and evaluation and analytical development. The presentation closes with a brief discussion of future plans.
SUMMARY OF TECHNOLOGY DEVELOPMENT

SEAL DEVELOPMENT:
- ADVANCED CONCEPT DEVELOPMENT
- FABRICATION TECHNOLOGY

ANALYTICAL DEVELOPMENT:
- SEAL LEAKAGE FLOW MODELING
- THERMAL-STRUCTURAL ANALYSES

EXPERIMENTAL DEVELOPMENT AND EVALUATION:
- HIGH TEMPERATURE LEAKAGE PERFORMANCE EVALUATION
- HIGH TEMPERATURE FRICTION AND WEAR ASSESSMENTS
  SOLID FILM LUBRICANT DEVELOPMENT

SCHEMATIC OF HIGH-TEMPERATURE PANEL-EDGE SEAL RIG (U)
UNCLASSIFIED

SEAL TEST FIXTURE
THERMAL EXPANSION (U)

- 0.5" rig expansion at temperature (enlarged for clarity)
- Surface conduction heaters (top and bottom)
- In-line air heaters (electric resistance)
- Seal test rig
- Air flow

UNCLASSIFIED

CERAMIC WAFER SEAL LEAKAGE MODEL (U)

- Movable engine panel
- P0
- H1
- H2
- Wafer
- M
- Ps
- g
- h1 and h2 enlarged for clarity

Where:

\[ M = \text{Seal leakage rate} \]
\[ h_1, h_2 = \text{Eff surface leakage gap} \]
\[ L = \text{Seal length} \]
\[ \mu (T) = \text{Power law viscosity} \]
\[ h_s = \frac{\Delta \text{CTE} x L x T}{N} = \text{small inter-wafer gap} \]
\[ N = \text{No. of interfaces} \]

\[ M = \frac{(P_s^2 - P_0^2)}{24 \mu R T} \left( \frac{L h_1^3}{H_1} + \frac{L h_2^3}{H_2} + \frac{N h_s^3}{H_2} \right) \]

Parasitic leakage around wafers
CTE mismatch induced leakage

ORIGINAL PAGE IS OF POOR QUALITY
COMPARISON OF
MEASURED AND PREDICTED LEAKAGE RATES VS TEMP.
FIXED PRESSURE DIFFERENTIAL: 40 psi

PRESSURE INDUCED WAFER LOADS

PRESSURE DISTRIBUTION:

FORCE BALANCE:

\[ \sum F_x = \int_0^H (p_2 - p_1) \, dx \]

\[ \sum F_y = \int_0^H (p_1 - p_2) \, dy + \int_{H_1}^{H_2} (p_1 - p_2) \, dy \]

\[ \sum M = \int_0^H (p_2 - p_1) \, H_2 \, dx + \int_0^H (p_1 - p_2) \, H_1 \, dy + \int_{H_1}^{H_2} (p_2 - p_1) \, H_1 \, dy \]

Pressure profiles:

\[ P_1(y) = \left( p_0 + \frac{(p_2 - p_1)}{H_2} (H_1 - y) \right)^{1/2} \]

\[ P_2(x) = \left( p_0 + \frac{(p_2 - p_1)}{H_2} (H_2 - x) \right)^{1/2} \]

Resultants:

\[ F_x = \left( p_2 - \frac{2}{3} \frac{(p_2 - p_1)}{p_o} \right) H_2 \]

\[ F_y = \left( p_2 - \frac{2}{3} \frac{(p_2 - p_1)}{p_o} \right) H_1 + (p_1 - p_2) g \]

\[ M = \frac{1}{2} \left( H_2^2 - H_1^2 \right) + \frac{2}{3} \frac{p_0^3}{(p_2 - p_1)^2} \left( H_2^2 - H_1^2 \right) - \frac{4}{15} \frac{(p_2 - p_1)^5}{(p_2 - p_0)^2} \left( H_2^4 - H_1^4 \right) \]

\[ \frac{(p_1 - p_2)}{g} \left( 2H_1 g + g^3 \right) \]
2-D BRAIDED ROPE SEAL LEAKAGE PERFORMANCE (U)
ROOM TEMPERATURE AIR, 80 PSI PRELOAD

Tentative leakage limit

\[ \Delta P = 60 \text{ psi} \]

\[ \text{Mass flow (lb/ft.-sec.)} \]

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<thead>
<tr>
<th>0</th>
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<tr>
<td>A-1</td>
<td>45 35</td>
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<td>B-1</td>
<td>30 35</td>
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<td>C-1</td>
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<td>D-1</td>
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<td>45 75</td>
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<tr>
<td>H-1</td>
<td>30 75</td>
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</tbody>
</table>

\[ \text{Movable engine panel - Splitter wall} \]

\[ \text{M} = \frac{P_s^2 - P_o^2}{L} \]

\[ \frac{1}{R} = \left( \text{Eff Seal Resistance} \right)^{-1} = \frac{1}{R_1} + \frac{1}{R_2} + \frac{1}{R_3} + \frac{1}{R_4} + \frac{1}{R_5 + R_6 + R_7} \]

Where:

- \( \frac{M}{L} = \text{Seal leakage per unit length} \)
- \( R_1, R_2 = \text{Resistance to flow behind and in-front of seal} \)
- \( R_3, R_4 = \text{Sheath resistance (eg parallel to flow direction)} \)
- \( R_5, R_7 = \text{Sheath resistance (eg perpendicular to flow direction, upstream and downstream)} \)
- \( R_6 = \text{Core resistance} \)
UNCLASSIFIED

BRAIDED ROPE SEAL LEAKAGE MEASURED AND PREDICTED (U)

Sample A1
Room temperature air, 80 psi preload

Sample G1
Room temperature air, 80 psi preload

Flow rate (lb/s-ft)

Pressure drop (psi)

SEAL DEVELOPMENT FUTURE PLANS

- EVALUATE ALTERNATE FIBER TOW DURABILITY OVER SIMULATED TEMPERATURES.

- EVALUATE COMBINED DURABILITY/LEAKAGE PERFORMANCE OF ADVANCED BRAID ARCHITECTURES UNDER ENGINE SIMULATED SLIDING CONDITIONS, TEMPERATURES AND PRESSURES.

- EVALUATE SURVIVABILITY OF BRAIDED ROPE SEALS UNDER THE ENGINE SIMULATED EROSIIVE SUPERSONIC FLOW FIELD.

- EXPERIMENTALLY ASSESS REQUIRED TRANSPERSIONATION COOLANT FLOW RATES TO SURVIVE THE HIGH ENGINE-SIMULATED HEAT FLUXES.
Mr. Cochrane discusses how advanced computer graphics visualization techniques and integration of knowledge throughout the concept to production cycle are providing tools to revolutionize the design process. Relevant projects that are used as examples are at USAF, Ford, Digital, EPA and DOE.
A mathematical model has been developed by BHR Group (formerly the British Hydromechanics Research Association) for the European Space Agency. The model couples the rigid body dynamics in four degrees of freedom of a flexibly-mounted stator (with Coulomb damping) of a turbopump face seal with the fluid film behavior. The interfacial film is assumed gaseous, and the model incorporates laminar/turbulent flow, inertia and sonic choking. Seal faces may be wavy and coned, whilst the rotating ring can be eccentrically mounted, misaligned with the shaft and oscillate axially at arbitrary frequency. The model is currently being coded into software. An experimental programme using a high speed cryogenic/hot gas test rig is being conducted to verify the output of the computer program. Future work planned for later in the year will incorporate mechanical contact effects.
APPENDIX A
WORKSHOP AGENDA

INTRODUCTION
Anita Liang
9:00-9:30

NASA/MTI CONTRACT SPECIFICS
MTI Industrial Codes
Wilbur Shapiro
9:30-9:50
CFDRC Scientific Codes
Andrzej Przekwas
9:50-10:10
KBS Development
Bharat Aggarwal
10:10-10:30

BREAK
10:30-10:45

DEMONSTRATION OF CYLINDRICAL AND SPIRAL-GROOVE INDUSTRIAL CODES
MTI
10:45-12:00

LUNCH (Main Cafeteria)
12:00-1:00

NASA SPONSORED ACTIVITIES
LeRC Related Activities
Bob Hendricks
1:00-1:15
MSFC Related Activities
Ted Benjamin
1:15-1:30
Army Related Activities
Bob Bill
1:30-1:40
Texas A&M Activities
Gerald Morrison
1:40-1:50
Akron University Activities
Jack Braun
1:50-2:00

PRESENTATION BY PEER REVIEW MEMBERS
Programs for Gas Turbine Engine Mainshaft Air/Oil Seals (AFSC) Ronald Dayton
2:00-2:10
Allison's Film Riding Face Seal and Brush Seal Work John Munson
2:10-2:20
Overview of GE's Seals Programs Nelson Pope
2:20-2:30
EG&G Fluid Components Technology Group Capabilities Harold Greiner
2:30-2:40
Application of Differential Roughness for Wear and Friction Alan Lebeck
2:40-2:50
Rocketdyne Fluid Seals/Bearing Joe Scharrer
2:50-3:00
Aerojet Activities George Bache
3:00-3:10
P&W Activities Saadat Syed
3:10-3:20
CFD Applications in Propulsion Charles Merkle
3:20-3:30

OPEN PRESENTATION
Cross Manufacturing's Brush Seal Ralph Flower
3:30-3:40
Activities

Seal/Bearing Rotordynamics Test Facility at CRUW Mike Adams
3:40-3:50
Labyrinth Seals Work at MIT Knox Millsaps
3:50-4:00
High T NASP Engine Seals Bruce Steinetz
4:00-4:10
Computer Interfaces for Knowledge Integration Tom Cochrane
4:10-4:20

ADJOURNMENT
Anita Liang
4:30
<table>
<thead>
<tr>
<th>Name</th>
<th>Organization/Institution</th>
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<tbody>
<tr>
<td>Mike Adams</td>
<td>Case Western Reserve University</td>
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<tr>
<td>Bharat Aggarwal</td>
<td>Mechanical Technology, Inc.</td>
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<tr>
<td>Mahesh Athavale</td>
<td>CFD Research Corp.</td>
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<td>Carlos Bailey</td>
<td>Turbo Components Textron</td>
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<td>Marc Carpino</td>
<td>Penn Statue University</td>
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<td>Tom Cochrane</td>
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<td>Ron Dayton</td>
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<td>F. K. Choy</td>
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<td>Ray Chupp</td>
<td>Teledyne CAE</td>
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<td>Rocketdyne</td>
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<td>John Kocur</td>
<td>Pratt &amp; Whitney</td>
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Joe Veres  
NASA LeRC

Jim Walker  
NASA LeRC

David Westcott  
Pratt & Whitney

Mark Wilson  
Allison Turbine Division

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In recognition of a deficiency in the current modeling capability for seals, a seven-year effort was established in 1990 by NASA's Office of Aeronautics, Exploration, and Technology under the Earth-to-Orbit Propulsion program of the Civilian Space Technology Initiative to develop verified computational-fluid-dynamic (CFD) concepts, codes, and analyses for seals. The objectives were to develop advanced concepts for the design and analysis of seals, to effectively disseminate the information to potential users by way of annual workshops, and to provide experimental verification for the models and codes under a wide range of operating conditions. The first annual workshop on "Seals Flow Code Development" was conducted at the NASA Lewis Research Center in Cleveland, Ohio, on March 26, 1991. The purpose was to report the progress of this program and to provide a forum for information exchange with the potential end-users. The presented material is documented in the NASA Proceedings.