NASA PATENT ABSTRACTS
BIBLIOGRAPHY

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SECTION 1 ABSTRACTS
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INTRODUCTION

Several thousand inventions result each year from the aeronautical and space research supported by the National Aeronautics and Space Administration. The inventions having important use in government programs or significant commercial potential are usually patented by NASA. These inventions cover practically all fields of technology and include many that have useful and valuable commercial application.

NASA inventions best serve the interests of the United States when their benefits are available to the public. In many instances, the granting of nonexclusive or exclusive licenses for the practice of these inventions may assist in the accomplishment of this objective. This bibliography is published as a service to companies, firms, and individuals seeking new, licensable products for the commercial market.

The NASA Patent Abstracts Bibliography (NASA PAB) is a semiannual NASA publication containing comprehensive abstracts and indexes of NASA-owned inventions covered by U.S. patents and applications for patent. The citations included in NASA PAB were originally published in NASA's Scientific and Technical Aerospace Reports (STAR) and cover STAR announcements made since May 1969.

For the convenience of the user, each issue of NASA PAB has a separately bound Abstract Section (Section 1) and Index Section (Section 2). Although each Abstract Section covers only the indicated six-month period, the Index Section is cumulative covering all NASA-owned inventions announced in STAR since 1969. Thus a complete set of NASA PAB would consist of the Abstract Sections of Issue 04 (January 1974) and Issue 12 (January 1978) and the Abstract Section for all subsequent issues and the Index Section for the most recent issue.


ABSTRACT SECTION (SECTION 1)

This PAB issue includes 10 major subject divisions separated into 76 specific categories and one general category/division. (See Table of Contents for the scope note of each category, under which are grouped appropriate NASA inventions.) This scheme was devised in 1975 and revised in 1987 in lieu of the 34 category divisions which were utilized in PAB supplements (01) through (06) covering STAR abstracts from May 1969 through January 1974. Each entry in the Abstract Section consists of a STAR citation accompanied by an abstract and, when appropriate, a key illustration taken from the patent or application for patent. Entries are arranged by subject category in order of the ascending NASA Accession Number originally assigned for STAR to the invention. The range of NASA Accession Numbers within each issue is printed on the inside front cover.

Abstract Citation Data Elements: Each of the abstract citations has several data elements useful for identification and indexing purposes, as follows:

- NASA Accession Number
- NASA Case Number
- Inventor's Name
- Title of Invention
- U.S. Patent Application Serial Number
- U.S. Patent Number (for issued patents only)
- U.S. Patent Office Classification Number(s)
  (for issued patents only)

These data elements are identified in the Typical Citation and Abstract and in the indexes.
INDEX SECTION (SECTION 2)

The Index Section is divided into five indexes. These indexes are cross-indexed and are used to locate a single invention or groups of inventions.

**_subject_index:** Lists all inventions according to appropriate alphabetized technical term and indicates the related NASA Case Number, the Subject Category Number, and the Accession Number.

**Inventor Index:** Lists all inventions according to alphabetized names of inventors and indicates the related NASA Case Number, the Subject Category Number, and the Accession Number.

**Source Index:** Lists all inventions according to alphabetized source of invention (i.e., name of contractor or government installation where invention was made) and indicates the related NASA Case Number, the Subject Category Number, and the Accession Number.

**Number Index:** Lists inventions in order of ascending (1) NASA Case Number, (2) U.S. Patent Application Serial Number, (3) U.S. Patent Classification Number, and (4) U.S. Patent Number and indicates the related Subject Category Number and the Accession Number.

**Accession Number Index:** Lists all inventions in order of ascending Accession Number and indicates the related Subject Category Number, the NASA Case Number, the U.S. Patent Application Serial Number, the U.S. Patent Classification Number, and the U.S. Patent Number.

**HOW TO USE THIS PUBLICATION TO IDENTIFY NASA INVENTIONS**

To identify one or more NASA inventions within a specific technical field or subject, several techniques are possible with the flexibility incorporated into the NASA PAB.

1. **Using Subject Category:** To identify all NASA inventions in any one of the subject categories in this issue of NASA PAB, select the desired Subject Category in the Abstract Section (Section 1) and find the inventions abstracted thereunder.

2. **Using Subject Index:** To identify all NASA inventions listed under a desired technical subject index term, (A) turn to the cumulative Subject Index in the Index Section and find the invention(s) listed under the desired technical subject term. (B) Note the indicated Accession Number and the Subject Category Number. (C) Using the indicated Accession Number, turn to the inside front cover of the Index Section to determine which issue of the Abstract Section includes the Accession Number desired. (D) To find the abstract of the particular invention in the issue of the Abstract Section selected, (1) use the Subject Category Number to locate the Subject Category and (2) use the Accession Number to locate the desired invention within the Subject Category listing.

3. **Using Patent Classification Index:** To identify all inventions covered by issued NASA patents (not including applications for patent) within a desired Patent Classification, (A) turn to the Patent Classification Number in the Number Index of Section 2 and find the associated invention(s), and (B) follow the instructions outlined in (2)(B), and (D) above.
A porous body of fibrous, low density silica-based insulation material is at least in part impregnated with a reactive boron oxide containing borosilicate glass frit, a silicon tetraboride fluxing agent and a molybdenum silicide emittance agent. The glass frit, fluxing agent and emittance agent are separately milled to reduce their particle size, then mixed together to produce a slurry in ethanol. The slurry is then applied to the insulation material and sintered to produce the porous body.

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Section 1 • Abstracts

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01 AERONAUTICS (GENERAL) ........................................................................... N.A.

02 AERODYNAMICS .............................................................................................. 1
Includes aerodynamics of bodies, combinations, wings, rotors, and control surfaces; and internal flow in ducts and turbomachinery. For related information see also 34 Fluid Mechanics and Heat Transfer.

03 AIR TRANSPORTATION AND SAFETY .......................................................... N.A.
Includes passenger and cargo air transport operations; and aircraft accidents. For related information see also 16 Space Transportation and 85 Urban Technology and Transportation.

04 AIRCRAFT COMMUNICATIONS AND NAVIGATION .................................. N.A.
Includes digital and voice communication with aircraft; air navigation systems (satellite and ground based); and air traffic control. For related information see also 17 Space Communications, Spacecraft Communications, Command and Tracking and 32 Communications and Radar.

05 AIRCRAFT DESIGN, TESTING AND PERFORMANCE ................................... 1
Includes aircraft simulation technology. For related information see also 18 Spacecraft Design, Testing and Performance and 39 Structural Mechanics. For land transportation vehicles see 85 Urban Technology and Transportation.

06 AIRCRAFT INSTRUMENTATION .................................................................... N.A.
Includes cockpit and cabin display devices; and flight instruments. For related information see also 19 Spacecraft Instrumentation and 35 Instrumentation and Photography.

07 AIRCRAFT PROPULSION AND POWER ......................................................... N.A.
Includes prime propulsion systems and systems components, e.g., gas turbine engines and compressors; and onboard auxiliary power plants for aircraft. For related information see also 20 Spacecraft Propulsion and Power, 28 Propellants and Fuels, and 44 Energy Production and Conversion.

08 AIRCRAFT STABILITY AND CONTROL ...................................................... N.A.
Includes aircraft handling qualities; piloting; flight controls; and autopilots. For related information see also 05 Aircraft Design, Testing and Performance.

09 RESEARCH AND SUPPORT FACILITIES (AIR) ......................................... N.A.
Includes airports, hangars and runways; aircraft repair and overhaul facilities; wind tunnels; shock tubes; and aircraft engine test stands. For related information see also 14 Ground Support Systems and Facilities (Space).

ASTRONAUTICS For related information see also Aeronautics.

12 ASTRONAUTICS (GENERAL) ........................................................................ N.A.
For extraterrestrial exploration see 91 Lunar and Planetary Exploration.

13 ASTRODYNAMICS ........................................................................................ N.A.
Includes powered and free-flight trajectories; and orbital and launching dynamics.

14 GROUND SUPPORT SYSTEMS AND FACILITIES (SPACE) ......................... 2
Includes launch complexes, research and production facilities; ground support equipment, e.g., mobile transporters; and simulators. For related information see also 09 Research and Support Facilities (Air).

15 LAUNCH VEHICLES AND SPACE VEHICLES .............................................. N.A.
Includes boosters; operating problems of launch/space vehicle systems; and reusable vehicles. For related information see also 20 Spacecraft Propulsion and Power.

16 SPACE TRANSPORTATION ............................................................................. 2
Includes passenger and cargo space transportation, e.g., shuttle operations; and space rescue techniques. For related information see also 03 Air Transportation and Safety and 18 Spacecraft Design, Testing and Performance. For space suits see 54 Man/Space Technology and Life Support.

17 SPACE COMMUNICATIONS, SPACECRAFT COMMUNICATIONS, COMMAND AND TRACKING .......................................................... N.A.
Includes telemetry; space communications networks; astronavigation and guidance; and radio blackout. For related information see also 04 Aircraft Communications and Navigation and 32 Communications and Radar.

N.A.—no abstracts were assigned to this category for this issue.
18  SPACECRAFT DESIGN, TESTING AND PERFORMANCE  ........................................ 3
Includes satellites; space platforms; space stations; spacecraft systems and components such as thermal and environmental controls; and attitude controls. For related information see 54 Man/System Technology and Life Support. For related information see also 05 Aircraft Design, Testing and Performance, 39 Structural Mechanics, and 16 Space Transportation.

19  SPACECRAFT INSTRUMENTATION .................................................................  N.A.
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20  SPACECRAFT PROPULSION AND POWER ................................................... 4
Includes main propulsion systems and components, e.g., rocket engines; and spacecraft auxiliary power sources. For related information see also 07 Aircraft Propulsion and Power, 28 Propellants and Fuels, 44 Energy Production and Conversion, and 15 Launch Vehicles and Space Vehicles.

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25  INORGANIC AND PHYSICAL CHEMISTRY .............................................. 7
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26  METALLIC MATERIALS .................................................................  N.A.
Includes physical, chemical, and mechanical properties of metals, e.g., corrosion; and metallurgy.

27  NONMETALLIC MATERIALS ................................................................. 9
Includes physical, chemical, and mechanical properties of plastics, elastomers, lubricants, polymers, textiles, adhesives, and ceramic materials. For composite materials see 24 Composite Materials.

28  PROPPELLANTS AND FUELS .................................................................  N.A.
Includes rocket propellants, igniters and oxidizers; their storage and handling procedures; and aircraft fuels. For related information see also 07 Aircraft Propulsion and Power, 20 Spacecraft Propulsion and Power, and 44 Energy Production and Conversion.

29  MATERIALS PROCESSING .................................................................  N.A.
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ENGINEERING For related information see also Physics.

31  ENGINEERING (GENERAL) ................................................................. 12
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32  COMMUNICATIONS AND RADAR .......................................................... 15
Includes radar; land and global communications; communications theory; and optical communications. For related information see also 04 Aircraft Communications and Navigation and 17 Space Communications, Spacecraft Communications, Command and Tracking. For search and rescue see 03 Air Transportation and Safety, and 16 Space Transportation.

33  ELECTRONICS AND ELECTRICAL ENGINEERING .................................. 17
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34  FLUID MECHANICS AND HEAT TRANSFER ........................................... 20
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<td>MECHANICAL ENGINEERING</td>
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<td>QUALITY ASSURANCE AND RELIABILITY</td>
<td>Includes product sampling procedures and techniques; and quality control.</td>
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<td>STRUCTURAL MECHANICS</td>
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<td>ENERGY PRODUCTION AND CONVERSION</td>
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<td>SPACE BIOLOGY</td>
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<td>62</td>
<td>COMPUTER SYSTEMS</td>
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63 CYBERNETICS
Includes feedback and control theory, artificial intelligence, robotics and expert systems. For related information see also 54 Man/System Technology and Life Support.

64 NUMERICAL ANALYSIS
Includes iteration, difference equations, and numerical approximation.

65 STATISTICS AND PROBABILITY
Includes data sampling and smoothing; Monte Carlo method; and stochastic processes.

66 SYSTEMS ANALYSIS
Includes mathematical modeling; network analysis; and operations research.

67 THEORETICAL MATHEMATICS
Includes topology and number theory.

PHYSICS For related information see also Engineering.

70 PHYSICS (GENERAL)
For precision time and time interval (PTTI) see 35 Instrumentation and Photography; for geophysics, astrophysics or solar physics see 46 Geophysics, 90 Astrophysics, or 92 Solar Physics.

71 ACOUSTICS
Includes sound generation, transmission, and attenuation. For noise pollution see 45 Environment Pollution.

72 ATOMIC AND MOLECULAR PHYSICS
Includes atomic structure, electron properties, and molecular spectra.

73 NUCLEAR AND HIGH-ENERGY PHYSICS
Includes elementary and nuclear particles; and reactor theory. For space radiation see 93 Space Radiation.

74 OPTICS
Includes light phenomena and optical devices. For lasers see 36 Lasers and Masers.

75 PLASMA PHYSICS
Includes magnetohydrodynamics and plasma fusion. For ionospheric plasmas see 46 Geophysics. For space plasmas see 90 Astrophysics.

76 SOLID-STATE PHYSICS
Includes superconductivity. For related information see also 33 Electronics and Electrical Engineering and 36 Lasers and Masers.

77 THERMODYNAMICS AND STATISTICAL PHYSICS
Includes quantum mechanics; theoretical physics; and Bose and Fermi statistics. For related information see also 25 Inorganic and Physical Chemistry and 34 Fluid Mechanics and Heat Transfer.

SOCIAL SCIENCES

80 SOCIAL SCIENCES (GENERAL)
Includes educational matters.

81 ADMINISTRATION AND MANAGEMENT
Includes management planning and research.

82 DOCUMENTATION AND INFORMATION SCIENCE
Includes information management; information storage and retrieval technology; technical writing; graphic arts; and micrography. For computer documentation see 61 Computer Programming and Software.

83 ECONOMICS AND COST ANALYSIS
Includes cost effectiveness studies.

84 LAW, POLITICAL SCIENCE AND SPACE POLICY
Includes NASA appropriation hearings; aviation law; space law and policy; international law; international cooperation; and patent policy.

85 URBAN TECHNOLOGY AND TRANSPORTATION
Includes applications of space technology to urban problems; technology transfer; technology assessment; and surface and mass transportation. For related information see 03 Air Transportation and Safety, 16 Space Transportation, and 44 Energy Production and Conversion.
SPACE SCIENCES For related information see also Geosciences.

88 SPACE SCIENCES (GENERAL) ......................................................... N.A.

89 ASTRONOMY ............................................................................. N.A.
Includes radio, gamma-ray, and infrared astronomy; and astrometry.

90 ASTROPHYSICS ........................................................................ N.A.
Includes cosmology; celestial mechanics; space plasmas; and interstellar and interplanetary gases and dust.
For related information see also 75 Plasma Physics.

91 LUNAR AND PLANETARY EXPLORATION ................................. N.A.
Includes planetology; and manned and unmanned flights. For spacecraft design or space stations see 18 Spacecraft Design, Testing and Performance.

92 SOLAR PHYSICS ........................................................................ N.A.
Includes solar activity, solar flares, solar radiation and sunspots. For related information see 93 Space Radiation.

93 SPACE RADIATION ..................................................................... N.A.
Includes cosmic radiation; and inner and outer earth's radiation belts. For biological effects of radiation see 52 Aerospace Medicine. For theory see 73 Nuclear and High-Energy Physics.

GENERAL
Includes aeronautical, astronautical, and space science related histories, biographies, and pertinent reports too broad for categorization; histories or broad overviews of NASA programs.

99 GENERAL .................................................................................. N.A.

Section 2 • Indexes

SUBJECT INDEX  CONTRACT NUMBER INDEX
INVENTOR INDEX  NUMBER INDEX
SOURCE INDEX  ACCESSION NUMBER INDEX
AERODYNAMICS

Includes aerodynamics of bodies, combinations, wings, rotors, and control surfaces; and internal flow in ducts and turbomachinery.

A method is provided for visualizing aerodynamic flow effects on a test surface. First, discrete quantities of a sublimating chemical such as naphthalene are distinctively colored via appropriate dyes or paints. Next, a uniform layer of the sublimating chemical having a particular color is applied to the test surface. This layer is covered with a second uniform layer of a different colored sublimating chemical, and so on until a composite of multi-colored layers is formed having a discrete thickness. Friction caused by an airflow results in the distinctly colored layers being removed in proportion to such aerodynamic flow characteristics as velocity and temperature, resulting in a multi-colored portrait which approximates the air flow on the underlying test surface.

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AIRCRAFT DESIGN, TESTING AND PERFORMANCE

Includes aircraft simulation technology.

An improvement in the lift and drag characteristics of a lifting surface is achieved by attaching a serrated panel to the trailing edge of the lifting surface. The serrations may have a saw-tooth configuration, with a 60 degree included angle between adjacent serrations. The serrations may vary in shape and size over the span-wise length of the lifting surface, and may be positioned at fixed or adjustable deflections relative to the chord of the lifting surface.

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14 GROUND SUPPORT SYSTEMS AND FACILITIES (SPACE)

Includes launch complexes, research and production facilities; ground support equipment, e.g., mobile transporters; and simulators.

N92-15081* National Aeronautics and Space Administration. Marshall Space Flight Center, Huntsville, AL.

ELECTROMAGNETIC MEISSNER EFFECT LAUNCHER Patent
GLEN A. ROBERTSON, inventor (to NASA) 21 May 1991
9 p Filed 31 Oct. 1989
(NASA-CASE-MFS-28323-1; US-PATENT-5,017,549;
US-PATENT-CLASS-244-63; INT-PATENT-CLASS-F41B-6/00)
Avail: US Patent and Trademark Office CSCL 14B

An electromagnetic projectile launcher provides acceleration of a superconducting projectile through the diamagnetic repulsion of the superconducting projectile. A superconducting layer is provided aft of the projectile, either directly on the projectile or on a platform upon which the projectile is carried, and a traveling magnetic field is caused to propagate along a magnetic field drive coil in which the projectile is disposed. The resulting diamagnetic repulsion between the superconducting projectile and the traveling magnetic field causes the projectile to be propelled along the coil. In one embodiment, a segmented drive coil is used to generate the traveling magnetic field.

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16 SPACE TRANSPORTATION

Includes passenger and cargo space transportation, e.g., shuttle operations; and space rescue techniques.

N92-10035* National Aeronautics and Space Administration. Langley Research Center, Hampton, VA.

SHUTTLE ORBITER WITH TELESCOPING MAIN PROPULSION UNIT AND PAYLOAD Patent Application
IAN O. MACCONOCHIE, inventor (to NASA) 7 Aug. 1991 12 p
(NASA-CASE-LAR-13586-1; NAS 1.71:LAR-13586-1;
US-PATENT-APPL-SN-743469) Avail: NTIS HC/MF A03 CSCL 22B

An improved Space Shuttle with variable internal volume is provided. The Space Shuttle Orbiter includes a telescoping main propulsion unit. This main propulsion unit contains the main rocket engines and fuel tanks and telescopes into the Space Shuttle. A variable cavity is located between this unit and the crew compartment. Accordingly, the positioning of the telescoping main propulsion unit determines the volume of the variable cavity. Thus, the volume of the variable length of the entire Space Shuttle may be increased or decreased to achieve desired configurations for optimal storage. In one embodiment of the invention, the payload also telescopes within the variable cavity.

NASA
predictable compact energy absorber for safely halting the cover plate as the retainers are extended as the net is deployed. The invention further includes a block of an energy-absorbing material positioned in the net for receiving loose debris produced by the explosive release of the cover plate.

Official Gazette of the U.S. Patent and Trademark Office

18 SPACECRAFT DESIGN, TESTING AND PERFORMANCE

Includes satellites; space platforms; space stations; spacecraft systems and components such as thermal and environmental controls; and attitude controls.

N92-15114* National Aeronautics and Space Administration. Lyndon B. Johnson Space Center, Houston, TX.
HYPERVERSITY IMPACT SHIELD Patent

A hypervelocity impact shield and method for protecting a wall structure, such as a spacecraft wall, from impact with particles of debris having densities of about 2.7 g/cu cm and impact velocities up to 16 km/s are disclosed. The shield comprises a stack of ultra thin sheets of impactor disrupting material supported and arranged by support means in spaced relationship to one another and mounted to cover the wall in a position for intercepting the particles. The sheets are of a number and spacing such that the impacting particle and the resulting particulates of the impacting particle and sheet material are successively impact-shocked to a thermal state of total melt and/or vaporization to a degree as precludes perforation of the wall. The ratio of individual sheet thickness to the theoretical diameter of particles of debris which may be of spherical form is in the range of 0.03 to 0.05. The spacing between adjacent sheets is such that the debris cloud plume of liquid and vapor resulting from an impacting particle penetrating a sheet does not puncture the next adjacent sheet prior to the arrival threat of fragment particulates of sheet material and the debris particle produced by a previous impact.

Official Gazette of the U.S. Patent and Trademark Office

N92-21999* National Aeronautics and Space Administration. Lyndon B. Johnson Space Center, Houston, TX.
ASSURED CREW RETURN VEHICLE Patent

A return vehicle is disclosed for use in returning a crew to Earth from low earth orbit in a safe and relatively cost effective manner. The return vehicle comprises a cylindrically-shaped crew compartment attached to the large diameter of a conical heat shield having a spherically rounded nose. On-board inertial navigation and cold gas control systems are used together with a de-orbit propulsion system to effect a landing near a preferred site on the surface of the Earth. State vectors and attitude data are loaded from the attached orbiting craft just prior to separation vehicle.

Official Gazette of the U.S. Patent and Trademark Office
SPACECRAFT PROPULSION AND POWER

Includes main propulsion systems and components, e.g., rocket engines; and spacecraft auxiliary power sources.

EXTENDED TEMPERATURE RANGE ROCKET INJECTOR

STEVEN J. SCHNEIDER, inventor (to NASA) 8 Oct. 1991

A rocket injector is provided with multiple sets of manifolds for supplying propellants to injector elements. Sensors transmit the temperatures of the propellants to a suitable controller which is operably connected to valves between these manifolds and propellant storage tanks. When cryogenic propellant temperatures are sensed, only a portion of the valves are opened to furnish propellants to some of the manifolds. When lower temperatures are sensed, additional valves are opened to furnish propellants to more of the manifolds.

Official Gazette of the U.S. Patent and Trademark Office

METHOD FOR PROVIDING REAL-TIME CONTROL OF A GASEOUS PROPELLANT ROCKET PROPULSION SYSTEM

BRIAN G. MORRIS, inventor (to NASA) 12 Nov. 1991

The new and improved methods and apparatus disclosed provide effective real-time management of a spacecraft rocket engine powered by gaseous propellants. Real-time measurements representative of the engine performance are compared with predetermined standards to selectively control the supply of propellants to the engine for optimizing its performance as well as efficiently managing the consumption of propellants. A priority system is provided for achieving effective real-time management of the propulsion system by first regulating the propellants to keep the engine operating at an efficient level and thereafter regulating the consumption ratio of the propellants. A lower priority level is provided to balance the consumption of the propellants so significant quantities of unexpended propellants will not be left over at the end of the scheduled mission of the engine.

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CHEMISTRY AND MATERIALS (GENERAL)

POLY 1, 2, 4-TRIAZOLES VIA AROMATIC NUCLEOPHILIC DISPLACEMENT Patent Application

The primary object of this invention constitutes new compositions of matter and a new process to prepare poly(1,2,4-triazoles) (PT). It concerns new PT, novel monomers, and the process for preparing the same. Another object of the present invention is to provide new PT that are useful as composite matrix resins for aircraft and dielectric interlayers in electronic devices. Another object of the present invention is the composition of several new di(hyproxyphenyl)-1,2,4-triazole monomers. According to the present invention, the foregoing and additional objects were obtained by synthesizing PT by the nucleophilic displacement reaction of di(hyproxyphenyl)-1,2,4-triazole monomers with activated aromatic dihalides. The inherent viscosities of the PT ranged from 1.37 to 3.4 dL/g and the glass transition temperatures ranged from 192 to 216 °C. One polymer exhibited a crystalline melting temperature of 377 °C. Thermogravimetric analysis (TGA) showed no weight loss occurring below 300 °C in air or nitrogen with a 5 percent weight loss occurring at approximately 500 °C in air and nitrogen. The synthesis of the
The di(hydroxphenyl)-1,2,4-triazole monomer is represented in an equation. The monomer can be prepared by either of the two routes shown. The chemistry can easily be extended to prepare similar di(hydroxphenyl)-1,2,4-triazole monomers as shown in a second equation. The aromatic dihydrazides in some cases are commercially available or readily prepared from hydrazine and a di(acid chloride). The substitution of the hydroxy groups in either type of monomer may be meta-meta, para-para, or para-meta. The general reaction sequence of PT from each type of di(hydroxyphenyl)-1,2,4-triazole monomer is shown in equations 3 and 4.

**N92-17882** National Aeronautics and Space Administration. Lewis Research Center, Cleveland, OH.

**SUBSTITUTED 1,1,1-TRIARYL-2,2,2-TRIFLUOROETHANES AND PROCESSES FOR THEIR SYNTHESIS Patent Application**

WILLIAM B. ALSTON, inventor (to NASA) and ROY F. GRATZ, inventor (to NASA) (Mary Washington Coll., Fredericksburg, VA.) 17 Jan. 1992 24 p


Synthetic procedures are given for tetraalkyl, tetraacid and dianhydrides substituted 1,1,1-triaryl-2,2,2-trifluoroethanes which comprises: (1) 1,1-bis (dialkylary) 1-aryl-2,2,2 trifluoroethane; (2) 1,1-bis (dicarboxyaryl) 1-aryl-2,2,2 trifluoroethane; or (3) cyclic dianhydride or diamine of 1,1-bis (dialkylaryl) 1-aryl-2,2,2 trifluoroethanes. The synthesis of (1) is accomplished by the condensation reaction of an aryltrifluoromethyl ketone with a dialkylaniline compound. The synthesis of (2) is accomplished by oxidation of (1). The synthesis dianhydride of (3) is accomplished by the conversion of (2) to its corresponding cyclic dianhydride. The synthesis of the diamine is accomplished by the similar reaction of an aryltrifluoromethyl ketone with aniline or alkyl substituted or disubstituted anilines. Also, other derivatives of the above are formed by nucleophilic displacement reactions.

**N92-16025** National Aeronautics and Space Administration. Lewis Research Center, Cleveland, OH.

**METHOD OF INTERCALATING LARGE QUANTITIES OF FIBROUS STRUCTURES Patent**

JAMES R. GAIER, inventor (to NASA) 17 Dec. 1991 7 p

Filed 2 Nov. 1990


A method of intercalating large quantities of fibrous structures uses a rotatable reaction chamber containing a liquid phase intercalate. The intercalate liquid phase is controlled by appropriately heating, cooling, or pressurizing the reaction. Rotation of the chamber containing the fiber sample enables total submergence of the fiber during intercalation. Intercalated graphite...
fibers having metal-like resistivities are achieved and are conceivably useful as electrical conductors.

Official Gazette of the U.S. Patent and Trademark Office

TOUGHENED UNI-PIECE FIBROUS INSULATION Patent

DANIEL B LEISER, inventor (to NASA), MARNELL SMITH, inventor (to NASA), REX A. CHURCHWARD, inventor (to NASA), and VICTOR W. KATVALA, inventor (to NASA) 7 Jan. 1992 10 p Filed 18 Jan. 1989


A porous body of fibrous, low density silica-based insulation material is at least in part impregnated with a reactive boron oxide containing borosilicate glass frit, a silicon tetraboride fluxing agent and a molybdenum silicide emittance agent. The glass frit, fluxing agent and emittance agent are separately milled to reduce their particle size, then mixed together to produce a slurry in ethanol. The slurry is then applied to the insulation material and sintered to produce the porous body.

Official Gazette of the U.S. Patent and Trademark Office

INTERCALATED HYBRID GRAPHITE FIBER COMPOSITE Patent Application

JAMES R. GAIER, inventor (to NASA) 25 Nov. 1991 13 p


The invention is directed to a highly conductive lightweight hybrid material and methods of producing the same. The hybrid composite is obtained by weaving strands of a high strength carbon or graphite fiber into a fabric-like structure, depositing a layer of carbon onto the structure, heat treating the structure to graphitize the carbon layer, and intercalating the graphitic carbon layer structure. A laminate composite material useful for protection against lightning strikes comprises at least one layer of the hybrid material over at least one layer of high strength carbon or graphite fibers. The composite material of the present invention is compatible with matrix compounds, has a coefficient of thermal expansion which is the same as underlying fiber layers, and is resistant to galvanic corrosion in addition to being highly conductive. These materials are useful in the aerospace industry, in particular as lightning strike protection for airplanes.

PRODUCTION OF MULLITE FIBERS Patent Application

DENNIS S. TUCKER, inventor (to NASA) and J. SCOTT SPARKS, inventor (to NASA) 23 Dec. 1991 15 p


Disclosed here is a process for making mullite fibers wherein a hydrolyzable silicon compound and an aluminum compound in the form of a difunctional aluminum chelate are hydrolyzed to form sols using water and an alcohol with a catalytic amount of hydrochloric acid. The sols are mixed in a molar ratio of aluminum to silicon of 3 to 1 and, under polycondensation conditions, a fibrous gel is formed. From this gel the mullite fibers can be produced.

NASA
25 INORGANIC AND PHYSICAL CHEMISTRY

Includes chemical analysis, e.g., chromatography; combustion theory; electrochemistry; and photochemistry.

N92-10073* National Aeronautics and Space Administration.
Pasadena Office, CA.
REGENERATIVE Cu/LA ZEOLITE SUPPORTED DESULFURIZING SORBENTS Patent

Efficient, regenerable sorbents for removal of H2S from fluid hydrocarbons such as diesel fuel at moderate condition comprise a porous, high surface area aluminosilicate support, suitably a synthetic zeolite, and most preferably a zeolite having a free lattice opening of at least 6 Angstroms containing from 0.1 to 0.5 moles of copper ions, lanthanum ions or their mixtures. The sorbent removes sulfur from the hydrocarbon fuel in high efficiency and can be repetitively regenerated without loss of activity.

Official Gazette of the U.S. Patent and Trademark Office

N92-12079* National Aeronautics and Space Administration.
Lyndon B. Johnson Space Center, Houston, TX.
METHOD FOR PRODUCING OXYGEN FROM LUNAR MATERIALS Patent Application

This invention is related to producing oxygen from lunar or Martian materials, particularly from lunar ilmenite in situ. The process includes producing a slurry of the minerals and hot sulfuric acid, the acid and minerals reacting to form sulfates of the metal. Water is added to the slurry to dissolve the minerals into an

N92-18561* National Aeronautics and Space Administration.
Pasadena Office, CA.
FABRICATION OF NANOMETER SINGLE CRYSTAL METALLIC COSI2 STRUCTURES ON Si Patent

Amorphous Co:Si (1:2 ratio) films are electron gun-evaporated on clean Si(111), such as in a molecular beam epitaxy system. These layers are then crystallized selectively with a focused electron beam to form very small crystalline Co/Si2 regions in an amorphous matrix. Finally, the amorphous regions are etched away selectively using plasma or chemical techniques.

Official Gazette of the U.S. Patent and Trademark Office

N92-21725* National Aeronautics and Space Administration.
Lewis Research Center, Cleveland, OH.
COMPPOSITE THERMAL BARRIER COATING Patent

A composite thermal barrier coating for a substrate is presented. The coating is comprised of a first layer that includes a ceramic material and a second layer that includes a ceramic material impregnated with glass. The glass is a ternary eutectic. The ceramic materials may include yttria-stabilized zirconia.

Official Gazette of the U.S. Patent and Trademark Office
aqueous solution, the first aqueous solution is separated from unreacted minerals from the slurry, and the aqueous solution is electrolyzed to produce the metal and oxygen.

NASA

fibers is released by heating to a temperature between 250 C and 400 C. The fluorine is then used to conduct fluorination reactions.

NASA

![Graph with data](image)

N92-19486* National Aeronautics and Space Administration. Lyndon B. Johnson Space Center, Houston, TX.

APPARATUS AND METHOD FOR CELLULOSE PROCESSING USING MICROWAVE PRETREATMENT Patent Application


A method for pretreating a cellulosic waste product with microwaves is disclosed as well as a method and apparatus for converting cellulosic waste into soluble saccharides. The invention greatly enhances a reaction rate for enzymatic hydrolysis. A feed mixture of cellulose, water, and acetic acid are irradiated with microwaves at a superatmospheric pressure in an autoclave reaction vessel and the treated cellulose is enzymatically hydrolyzed in a bioreactor. The acid and enzymes are optionally separated for reuse. As a feed stock for the culture of microbes, the sugars can be further processed into ethanol or food protein. High yield, low hazard potential, low energy usage and ready preparation in space of acetic acid and the enzyme makes the present invention well suited for use on long duration space missions.

NASA

![Patent diagram](image)

N92-17902* National Aeronautics and Space Administration. Lewis Research Center, Cleveland, OH.

STORING FLUORINE GAS IN CARBON FIBERS AND RELEASING THE SAME Patent Application


Fluorine gas stored in pitch-based graphitized carbon...
27  NONMETALLIC MATERIALS

Includes physical, chemical, and mechanical properties of plastics, elastomers, lubricants, polymers, textiles, adhesives, and ceramic materials.

N92-10090* National Aeronautics and Space Administration. Lewis Research Center, Cleveland, OH.

BROMINATED GRAPHITIZED CARBON FIBERS Patent
CHING-CHEH HUNG, inventor (to NASA) 22 Oct. 1991
83 p Filed 30 Nov. 1989 Supersedes N90-15262 (28 - 7, p 910)
Continuation-in-part of abandoned US-Patent-App-SN-219016, filed
14 Jul. 1988
(NASA-CASE-LEW-14698-2; US-PATENT-5,059,409;
INT-PATENT-CLASS-C01B-31/04) Avail: US Patent and
Trademark Office CSCL 11C

Low cost, high break elongation graphitized carbon fibers
having low degree of graphitization are inert to bromine at room
or higher temperatures, but are brominated at -7 to 20 C, and
then debrominated at ambient. Repetition of this
bromination-debromination process can bring the bromine content
to 18 percent. Electrical conductivity of the brominated fibers is
three times of the before-bromination value.

Official Gazette of the U.S. Patent and Trademark Office

N92-10091* National Aeronautics and Space Administration.
Lyndon B. Johnson Space Center, Houston, TX.

HELMET OF A LAMINATE CONSTRUCTION OF
POLYCARBONATE AND POLYSULFONE POLYMERIC
MATERIAL Patent
JOSEPH J. KOSMO, inventor (to NASA) and FREDERIC S.
DAWN, inventor (to NASA) 15 Oct. 1991 60 p Filed 30 Nov.
1989 Supersedes N90-16925 (28 - 9, p 1195)
(NASA-CASE-MSC-21503-1; US-PATENT-5,056,156;

An article of laminate construction is disclosed which is
comprised of an underlayer of polycarbonate polymer material to
which is applied a chemically resistant outer layer of polysulfone.
The layers which are joined by compression-heat molding, are
molded to form the shape of a body protective shell such as a
space helmet comprising a shell of polycarbonate, polysulfone
laminate construction attached at its open end to a sealing ring
adapted for connection to a space suit. The front portion of the
shell provides a transparent visor for the helmet. An outer visor
of polycarbonate polysulfone laminate construction is pivotally
mounted to the sealing ring for covering the transparent visor
portion of the shell during extravehicular activities. The
polycarbonate under layer of the outer visor is coated on its inner
surface with a vacuum deposit of gold to provide additional thermal
radiation resistance.

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N92-10105* National Aeronautics and Space Administration.
Langley Research Center, Hampton, VA.

A PROCESS FOR PREPARING 1,3-DIAMINO-5-
PENTAFLUOROSULFANYLBENZENE AND POLYMERS
THEREFROM Patent Application
ANNE K. ST.CLAIR, inventor (to NASA), TERRY L. ST.CLAIR,
inventor (to NASA), and JOSEPH S. THRASHER, inventor (to
(NASA-CASE-LAR-14773-1-CU; NAS 1.71:LAR-14773-1;
US-PATENT-APPL-SN-755207) Avail: NTIS HC/MF A03 CSCL
11C

Diamines have shown their utility in the formation of
many polymers. Examples of these polymers include polyimides,
polyamides, and epoxies. The properties of these polymers are
often dependent on the diamine which is used to make the polymer.
By the present invention, a process was developed to make a
diamine containing pentafluorosulfanylbenzene moiety. This
process involves two steps: the preparation of a dinitro precursor
and the reduction of the dinitro compound to form the diamine.
This diamine was then reacted with various dihydrides,
diacidchlorides, and epoxy resins to yield the corresponding
polyamide, polyamide, and epoxy polymers. These polymers were
then used to make films, a wire coating enamel, and a
semi-permeable membrane. The novelty of this invention resides
in the process to make the diamine. Traditionally, dinitro compounds
are reduced with hydrazine or a catalyst such as palladium on
The catalyst which is used in this invention is platinum oxide. When this catalyst is used, it makes it possible to form a polymer-grade diamine.

A solid lubricant film on a pretreated surface is described. The surface topography of the material to be lubricated is first selectively altered. Photochemical etching is employed to selectively determine contact area and shape to maximize the proper ratio of reservoir area to sliding contact area. Cadmium oxide is then sputtered onto the altered surface. The cadmium oxide acts as an intermediate layer to more tightly bond the solid lubricant, such as graphite, onto the material surface.

The semicrystalline polyamide prepared by reaction of 3,3',4,4'-benzophenonetetracarboxylic (BTDA) and 1,3-bis(4-aminophenoxy 4' benzoyl) benzene (1,3-BABB) is modified so that it can be more readily processed to form adhesive bonds, moldings, and composites. The stoichiometric ratio of the two monomers, BTDA and 1,3-BABB is controlled so that the intermediate polyamide acid is of a calculated molecular weight. A polyamide acid with excess anhydride groups is then reacted with the stoichiometrically required amount of monofunctional aromatic or aliphatic amine required for complete endcapping. The stoichiometrically offset, encapped polyamide is processed at lower temperatures and pressures than the unmodified high molecular weight polyamide with the same repeat unit, and exhibits an improved melt stability.
A high temperature stable, highly optically transparent to colorless, low dielectric linear aromatic polyamide is prepared by reacting an aromatic diamine with 3,ba'as (3,4-dicarboxyphenoxy) diphenylmethane dianhydride in an amide solvent to form a linear aromatic polyamic acid. This polyamic acid is then cyclized to form the corresponding polyamide. The general structural formula is given.

**Route for preparing PDMDA**

![Diagram of Route for preparing PDMDA](image)

**FIGURE 1**

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**N92-11201**


An assembly of an article and a polyamide is prepared. The assembly resists dimensional change, delamination, or debonding when exposed to changes in temperature. An article is provided. A soluble polyamide resin solution having a low coefficient of thermal expansion (CTE) was prepared by dissolving the polyamide in solvent and adding a metal ion-containing additive to the solution. Examples of the additive are: Ho(OOCCH3), Er(NPPA)3, TmCl3, and Er(C5H7O2)3. The soluble polyamide resin is combined with the article to form the assembly.

**FIGURE 2**

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**N92-16121**


An electropolymerized film comprised of polymers and copolymers of a monomer is formed on the surface of an anode. The finished structures have superior electrical and mechanical properties for use in applications such as electrostatic dissipation.
and for the reduction of the radar cross section of advanced aircraft.

**Official Gazette of the U.S. Patent and Trademark Office**

**N62-16122** National Aeronautics and Space Administration. Lewis Research Center, Cleveland, OH. **METHOD OF MAKING CONTAMINATION-FREE CERAMIC BODIES Patent**


Ceramic bodies having unique properties are provided by the present invention. These bodies are especially useful in a high temperature, high thermal conductivity application. The relatively low density and low thermal expansion of these bodies enhance their usage in advanced space and missile technology and aircraft components.

**Official Gazette of the U.S. Patent and Trademark Office**

**N62-16123** National Aeronautics and Space Administration. Marshall Space Flight Center, Huntsville, AL. **SPRAYABLE LIGHTWEIGHT ABLATIVE COATING Patent**


An improved lightweight, ablative coating is disclosed that may be spray applied and cured without the development of appreciable shrinkage cracks. The ablative mixture consists essentially of phenolic microballoons, hollow glass spheres, glass fibers, ground cork, a flexibilized resin binder, and an activated colloidal clay.

**Official Gazette of the U.S. Patent and Trademark Office**

**N62-17676** National Aeronautics and Space Administration. Langley Research Center, Hampton, VA. **POLYMIDES CONTAINING AMIDE AND PERFLUOROISOPROPYL CONNECTING GROUPS Patent Application**


New, thermooxidatively stable polyimides were prepared from the reaction of aromatic dianhydrides containing isopropylidene bridging groups with aromatic diamines containing amide connecting groups between the rings. Several of these polyimides were shown to be semi-crystalline as evidenced by wide angle x-ray scattering and differential scanning calorimetry. Most of the polyimides form tough, flexible films with high tensile properties. These polyamide films exhibit enhanced solubility in organic solvents.

**Official Gazette of the U.S. Patent and Trademark Office**

**N62-21711** National Aeronautics and Space Administration. Langley Research Center, Hampton, VA. **SEMI-INTERPENETRATING POLYMER NETWORK FOR TOUGHER AND MORE MICROCRACKING RESISTANT HIGH TEMPERATURE POLYMERS Patent**


This invention is a semi-interpenetrating polymer network which includes a high performance thermostetting polyamide having a nadic end group acting as a crosslinking site and a high performance linear thermoplastic polyamide. An improved high temperature matrix resin is provided which is capable of performing at 316 C in air for several hundreds of hours. This resin has significantly improved toughness and microcracking resistance, excellent processability and mechanical performance, and cost effectiveness.

**Official Gazette of the U.S. Patent and Trademark Office**

**N62-22044** National Aeronautics and Space Administration. Langley Research Center, Hampton, VA. **TOUGH, HIGH PERFORMANCE, ADDITION-TYPE THERMOPLASTIC POLYMERS Patent**


A tough, high performance polyamide is provided by reacting a trialetic bond conjugated with an aromatic ring in a bisacetyl compound with the active double bond in a compound containing a double bond activated toward the formation of a Diels-Adler type adduct, especially a bismaleimide, a bisbistraconimide, or a benzoquinone, or mixtures thereof. Addition curing of this product produces a high linear polymeric structure and heat treating the highly linear polymeric structure produces a thermally stable aromatic addition-type thermoplastic polyamide, which finds utility in the preparation of molding compounds, adhesive compositions, and polymer matrix composites.

**Official Gazette of the U.S. Patent and Trademark Office**

**31 ENGINEERING (GENERAL)**

Includes vacuum technology; control engineering; display engineering; cryogenics; and fire prevention.
INTEGRAL FILL YARN INSERTION AND BEATUP METHOD
Patent Application
GARY L. FARLEY, Inventor (to NASA) 26 Sep. 1991 14 p
(NASA-CASE-LAR-14046-1; NAS 1.71:LAR-14046-1;

An apparatus and method for integral fill yarn insertion and beatup are disclosed. A modified rapier contains a channel for holding fill yarn. The channel is covered with a flexible and inflatable boot, and an inflating apparatus for this boot is also attached. Fill yarn is inserted into the channel, and the rapier is extended into a shed formed by warp yarn. Next, the rapier is pushed into the fell of the fabric, and the flexible and inflatable cover inflated, which both pushes the yarn into the fell of the fabric and performs beatup. The rapier is withdrawn and the shed closed to complete one step of the weaving process.

N92-15203* National Aeronautics and Space Administration.
Pasadena Office, CA.
MULTICOMPONENT GAS SORPTION JOULE-THOMSON REFRIGERATION Patent
(NASA-CASE-NPO-17569-1-CU; US-PATENT-5,063,747;
US-PATENT-CLASS-624-3.2; US-PATENT-CLASS-624-51.2;

The present invention relates to acryogenic Joule-Thomson refrigeration capable of pumping multicomponent gases with a single stage sorption compressor system. Alternative methods of pumping a multicomponent gas with a single stage compressor are disclosed. In a first embodiment, the sorbent geometry is such that a void is defined near the output of the sorption compressor. When the sorbent is cooled, the sorbent primarily adsorbs the higher boiling point gas such that the lower boiling point gas passes through the sorbent to occupy the void. When the sorbent is heated, the higher boiling point gas is desorbed at high temperature and pressure and thereafter propels the lower boiling point gas out of the sorption compressor. A mixing chamber is provided to remix the constituent gases prior to expansion of the gas through a Joule-Thomson valve. Other methods of pumping a multicomponent gas are disclosed. For example, where the sorbent is porous and the low boiling point gas does not adsorb very well, the pores of the sorbent will act as a void space for the lower boiling point gas. Alternatively, a mixed sorbent may be used where a first sorbent component physically adsorbs the high boiling point gas and where the second sorbent component chemically absorbs the low boiling point gas.

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N92-16161* National Aeronautics and Space Administration.
Lyndon B. Johnson Space Center, Houston, TX.
METHOD AND APPARATUS FOR RELEASABLY CONNECTING FIRST AND SECOND OBJECTS Patent
(NASA-CASE-MSC-21517-1; US-PATENT-5,061,112;
The apparatus and method are disclosed for releasably connecting first and second objects, where a magnetic end effector may include at least one elongated pin number, a proximal end of which is connected to the first object and the distal end of which may be inserted into a receiving portion in the second object. Latch members are carried by the pin member for radial movement between retracted and expanded positions for releasing and locking, respectively, first and second objects. A plunger member carried by the pin member is axially moveable between first and second positions. In the first plunger position, the latch members are located in the expanded (locked) position and in the second plunger position the latch members are released for movement to retracted or unlocked position. The magnetic end effector is provided for releasable attachment to the first object and for moving the plunger member to the second position, releasing the first object.

The invention is an apparatus and method for wire splicing using an explosive joining process. The apparatus consists of a prebent, U-shaped strap of metal that slides over propositioned wires. A standoff means separates the wires from the strap before joining. An adhesive means holds two ribbon explosives in position centered over the U-shaped strap. A detonating means connects to the ribbon explosives that drive the strap to accomplish a high velocity, angular collision between the mating surfaces. This collision creates surface melts and collision bonding results in electron sharing linkups.

A three-stage sorption type cryogenic refrigeration system, each stage containing a fluid having a respectively different boiling point. Each stage includes a compressor in which a respective fluid is heated to be placed in a high pressure gaseous state. The compressor for that fluid which is heated to the highest temperature is enclosed by the other two compressors to permit heat to be transferred from the inner compressor to the surrounding compressors. The system may include two sets of compressors, each having the structure described above, with the interior compressors of the two sets coupled together to permit selective heat transfer therebetween, resulting in more efficient utilization of input power.
SHARPS CONTAINER Patent Application
ANGELENE M. LEE, inventor (to NASA) 7 Oct. 1991 13 p
(NASA-CASE-MSC-21776-1; NAS 1.71:MSC-21776-1;
US-PATENT-APPL-SN-772763) Avail: NTIS HC/MF A03 CSCL
13B

This invention relates to a system for use in disposing of potentially hazardous items and more particularly a Sharps receptacle for used hypodermic needles and the like. A Sharps container is constructed from lightweight alodined nonmagnetic metal material with a cup member having an elongated tapered shape and length greater than its transverse dimensions. A magnet in the cup member provides for metal retention in the container. A nonmagnetic lid member has an opening and spring biased closure flap member. The flap member is constructed from stainless steel. A Velcro patch on the container permits selective attachment at desired locations.

THE INVENTION

The invention relates to a method and apparatus for frequency spectrum analysis of an unknown signal in real time based upon integration of 1-bit samples of signal voltage amplitude corresponding to sine or cosine phases of a controlled center frequency clock which is changed after each integration interval to sweep the frequency range of interest in steps. Integration of samples during each interval is carried out over a number of cycles of the center frequency clock spanning a number of cycles of an input signal to be analyzed. The invention may be used to detect the frequency of at least two signals simultaneously. By using a reference signal of known frequency and voltage amplitude (added to the two signals for parallel processing in the same way, but in a different channel with a sampling at the known frequency and phases of the reference signal), the absolute voltage amplitude of the other two signals may be determined by squaring the sine and cosine integrals of each channel and summing the squares to obtain relative power measurements in all three channels and, from the known voltage amplitude of the reference signal, obtaining an absolute voltage measurement for the other two signals by multiplying the known voltage of the reference signal with the ratio of the relative power of each of the other two signals to the relative power of the reference signal.

COMUNICATIONS AND RADAR

Includes radar; land and global communications; communications theory; and optical communications.

C-17B

A system for real-time video image display for robotics or remote-vehicle teleoperation is described that has at least one robot arm or remotely operated vehicle controlled by an operator through hand-controllers, and one or more television cameras and optional lighting element. The system has at least one television monitor for display of a television image from a selected camera and the ability to select one of the cameras for image display. Graphics are generated with icons of cameras and lighting elements for display surrounding the television image to provide the operator information on: the location and orientation of each camera and lighting element; the region of illumination of each lighting element; the viewed region and range of focus of each camera; which
camera is currently selected for image display for each monitor; and when the controller coordinate for said robot arms or remotely operated vehicles have been transformed to correspond to coordinates of a selected or nonselected camera.

REAL-TIME DATA COMPRESSION OF BROADCAST VIDEO SIGNALS Patent
MARTY J. WALKER, inventor (to NASA) 15 Oct. 1991 30 p Filed 9 Nov. 1990
Supersedes N91-15469 (29-7, p 996) Continuation-in-part of

This is a fully parallel analog backpropagation learning processor which comprises a plurality of programmable resistive
memory elements serving as synapse connections whose values can be weighted during learning with buffer amplifiers, summing circuits, and sample-and-hold circuits arranged in a plurality of neuron layers in accordance with delta-backpropagation algorithms modified so as to control weight changes due to circuit drift.

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A method for fabricating self-aligned n-type guard rings in a body of p-type silicon for a silicide/Schottky-barrier diode array consists of depositing SiO2 with uniformly distributed phosphorus ions as a masking layer on the surface of the p-doped silicon body before etching the masking layer in the desired pattern of guard rings. N-type guard rings buried in the p-type silicon substrate are then formed by heating the structure to diffuse the phosphorus ions remaining on the surface functions as a mask for the remaining steps of the process which are to deposit metal over the entire surface of the structure and then form a silicide in those areas inside the SiO2 grid pattern by heating the structure. Metal not converted to a silicide is then removed by etching. The silicide and p-type silicon form a Schottky-barrier diode at their junction in areas inside the buried guard rings. Overlap of the silicide and the guard rings is minimized by this self-aligning technique to maximize the fill factor of the array.

N92-12174# National Aeronautics and Space Administration.
Langley Research Center, Hampton, VA.
VACUUM-ISOLATION VESSEL AND METHOD FOR MEASUREMENT OF THERMAL NOISE IN MICROPHONES Patent Application
ALLAN J. ZUCKERWAR, inventor (to NASA) and KIM CHI T. NGO, inventor (to NASA) (Old Dominion Univ., Norfolk, VA.)
7 Oct. 1991 17 p

The vacuum isolation vessel and method in accordance with the present invention are used to accurately measure thermal noise in microphones. The apparatus and method could be used in a microphone calibration facility or any facility used for testing microphones. Thermal noise is measured to determine the minimum detectable sound pressure by the microphone. Conventional isolation apparatus and methods have been unable to provide an acoustically quiet and substantially vibration free environment for accurately measuring thermal noise. In the present invention, an isolation vessel assembly comprises a vacuum sealed outer vessel, a vacuum sealed inner vessel, and an interior suspension assembly coupled between the outer and inner vessels for suspending the inner vessel within the outer vessel. A noise measurement system records thermal noise data from the isolation vessel assembly. A vacuum system creates a vacuum between an internal surface of the outer vessel and an external surface of the inner vessel. The present invention thus provides an acoustically quiet environment due to the vacuum created between the inner and outer vessels and a substantially vibration free environment due to the suspension assembly suspending the inner vessel within the outer vessel. The thermal noise in the microphone, effectively isolated according to the invention, can be accurately measured.

N92-10146# National Aeronautics and Space Administration.
Pasadena Office, CA.
FORMATION OF SELF-ALIGNED GUARD RINGS FOR MONOLITHIC SCHOTTKY-BARRIER DIODE ARRAYS Patent Application
TRUE-LON LIN, inventor (to NASA) (Jet Propulsion Lab., California Inst. of Tech., Pasadena.) 16 May 1991 16 p
(Contract NAS7-918)

A method for fabricating self-aligned n-type guard rings in a body of p-type silicon for a silicide/Schottky-barrier diode array consists of depositing SiO2 with uniformly distributed phosphorus ions as a masking layer on the surface of the p-doped silicon body before etching the masking layer in the desired pattern of guard rings. N-type guard rings buried in the p-type silicon substrate are then formed by heating the structure to diffuse the phosphorus ions remaining on the surface functions as a mask for the remaining steps of the process which are to deposit metal over the entire surface of the structure and then form a silicide in those areas inside the SiO2 grid pattern by heating the structure. Metal not converted to a silicide is then removed by etching. The silicide and p-type silicon form a Schottky-barrier diode at their junction in areas inside the buried guard rings. Overlap of the silicide and the guard rings is minimized by this self-aligning technique to maximize the fill factor of the array.
A magnetostrictive drive motor is disclosed which has a rotary drive shaft in the form of a drum which is encircled by a plurality of substantially equally spaced roller members in the form of two sets of cones which are in contact with the respective cam surfaces on the inside surface of an outer drive ring. The drive ring is attached to sets of opposing pairs of magnetostrictive rods. Each rod in a pair is mutually positioned end to end within respective energizing coils. When one of the coils in an opposing pair is energized, the energized rod expands while the other rod is caused to contract, causing the drive ring to rock, i.e., rotate slightly in either the clockwise or counterclockwise direction, depending upon which rod in a pair is energized. As the drive ring is activated in repetitive cycles in either direction, one set of drive cones attempts to roll up their respective cam surface but are pinned between the drive shaft drum and the drive ring. As the frictional force preventing sliding builds up, the cones become locked, setting up reaction forces including a tangential component which is imparted to the drive shaft drum to provide a source of motor torque. Simultaneously the other set of cones are disengaged from the drive shaft drum. Upon deactivation of the magnetostrictive rod coils, the force on the drive cones is released, causing the system to return to an initial rest position. By repetitively cycling the energization of the magnetostrictive rods, the drive shaft drum indexes in microradian rotational steps.

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A method for estimating the size and location of couplings within a waveguide directional coupler is provided. The method is applied to a waveguide directional coupler having a main transmission waveguide connected to an auxiliary transmission waveguide by a number of bore hold couplings. The bore hold couplings are in the interior of the waveguide directional coupler and, therefore, are not easily measurable. The method generally includes the steps of applying a two-sided tape to a member, inserting and securing the member within the main transmission waveguide, pouring a fine particulate substance such as talc into the auxiliary transmission waveguide such that a portion of the talc enters the bore hole couplings and adheres to the two-sided tape, and withdrawing the member such that the size and location of the bore hole couplings can be determined by measuring the size and location of marks on the two-sided tape caused by the fine particulate substance adhering to the two-sided tape.

A very high voltage amplifier is provided in which plural cascaded banks of capacitors are switched by optically isolated control switches so as to be charged in parallel from the preceding stage or capacitor bank and to discharge in series to the succeeding stage or capacitor bank in alternating control cycles. The optically isolated control switches are controlled by a logic controller whose power supply is virtually immune to interference from the very high voltage output of the amplifier by the optical isolation provided by the switches, so that a very high voltage amplification ratio may be attained using many capacitor banks in cascade.

The invention is a linear actuator that operates under the principle that like charges repel and opposite charges attract. The linear actuator consists of first and second pairs of spaced opposed conductors where one member of each pair of conductors is attached to a fixed member, and where the other member of each pair of conductors is attached to a movable member such as an elongated rod. The two pairs of spaced conductors may be provided in the form of two spacedly interwound helical vanes where the conductors are located on the opposite sides of the two helical vanes. One helical vane extends inwardly from a housing and the other helical vane extends outwardly from an elongated rod. The elongated rod may be caused to move linearly with respect...
to the housing by applying appropriate charges of like or opposite polarity to the electrical conductors on the helical vanes.

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FLUID MECHANICS AND HEAT TRANSFER

Includes boundary layers; hydrodynamics; fluidics; mass transfer; and ablation cooling.

N92-10167*# National Aeronautics and Space Administration. Lewis Research Center, Cleveland, OH.
PULSE THERMAL ENERGY TRANSPORT SYSTEM Patent Application

A pulse-thermal pump having a novel fluid flow wherein heat admitted to a closed system raises the pressure in a closed evaporator chamber, while another interconnected evaporator chamber remains open, is described. This creates a pressure differential, and at a predetermined pressure, the closed evaporator is opened and the opened evaporator is closed. The difference in pressure initiates fluid flow in the system.

N92-16241* National Aeronautics and Space Administration. Lyndon B. Johnson Space Center, Houston, TX.
HIGH VELOCITY GAS PARTICULATE SAMPLING SYSTEM Patent

A gas sampling system is disclosed for determining particulate matter contamination in a high velocity gas flow where the sampling chamber is first cleaned, then evacuated and is coupled by a closed three way valve in a straight line relationship to the gas supply line. A predetermined gas flow rate is established through the three way valve which is quickly opened to couple
the dynamically flowing gas to the evacuated sample chamber in a straight line relationship to trap a gas sample under dynamic conditions. When the sampling chamber has a gas sample, the three-way valve is again closed so that particulate matter in the sample chamber can be flushed from the sample chamber with a compatible liquid to a filter for collection and analysis.

Official Gazette of the U.S. Patent and Trademark Office


A flow reactor for simulating the interaction in the troposphere is set forth. A first reactant mixed with a carrier gas is delivered from a pump and flows through a duct having louvers therein. The louvers straighten out the flow, reduce turbulence and provide laminar flow discharge from the duct. A second reactant delivered from a source through a pump is input into the flowing stream, the second reactant being diffused through a plurality of small diffusion tubes to avoid disturbing the laminar flow. The commingled first and second reactants in the carrier gas are then directed along an elongated duct where the walls are spaced away from the flow of reactants to avoid wall interference, disturbance or turbulence arising from the walls. A probe connected with a measuring device can be inserted through various sampling ports in the second duct to complete measurements of the first and second reactants and the product of their reaction at selected XYZ locations relative to the flowing system.

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A device is provided for controlling pressure loading of a member caused by a fluid moving past the member or the member moving through a fluid. The device consists of two porous skins mounted over the solid surface of the member and separated from the solid surface by a plenum. Fluid from an area exerting high pressure on the member may enter the plenum through the pressure skins and exit into an area exerting a lower pressure on the member, thus controlling pressure loading of the member. The porous inner skin may be translated relative to the porous skin in order to achieve a porosity ranging from zero to the porosity of the porous skins.

NASA


A device is provided for controlling pressure loading of a member caused by a fluid moving past the member or the member moving through a fluid. The device consists of a porous skin mounted over the solid surface of the member and separated from the solid surface by a plenum. Fluid from an area exerting high pressure on the member may enter the plenum through the porous surface and exit into an area exerting a lower pressure on the member, thus controlling pressure loading of the member.

NASA
ADJUSTABLE STEAM PRODUCING FLEXIBLE ORIFICE INDEPENDENT OF FLUID PRESSURE Patent

Avail: US Patent and Trademark Office CSCL 20D

A self-adjusting choke for a fluids nozzle includes a membrane constructed of a single piece of flexible or elastic material. This flexible material is shaped to fit into the outlet of a nozzle. The body of the membrane has at least two flow channels, from one face to the other, which directs two streams of water to cross at the opening of the nozzle or at some point beyond. The elasticity and thickness of the membrane is selected to match the range of expected pressures and fluid velocities. The choke may have more than two flow channels, as long as they are aligned adjacent to one another and directed towards each other at the exit face. In a three orifice embodiment, one is directed upward, one is directed downward, and the one in the middle is directed forward. In this embodiment all three fluid streams intersect at some point past the nozzle opening. Under increased pressure the membrane will deform causing the orifices to realign in a more forward direction, causing the streams to intersect at a smaller angle. This reduces the force with which the separate streams impact each other, still allowing the separate streams to unify into a single stable spiralling stream in spite of the increased pressure.

INSTRUMENTATION AND PHOTOGRAPHY

Includes remote sensors; measuring instruments and gages; detectors; cameras and photographic supplies; and holography.

CONSTANT FREQUENCY PULSED PHASE-LOCKED LOOP MEASURING DEVICE Patent Application

N92-21724* National Aeronautics and Space Administration. Pasadena Office, CA.

N92-10185* National Aeronautics and Space Administration. Langley Research Center, Hampton, VA.

DUAL STRAIN GAGE BALANCE SYSTEM FOR MEASURING LIGHT LOADS Patent

Avail: US Patent and Trademark Office CSCL 14B

A dual strain gage balance system for measuring normal and axial forces and pitching moment of a metric airfoil model imparted by aerodynamic loads applied to the airfoil model during wind tunnel testing includes a pair of non-metric panels being rigidly connected to and extending towards each other from opposite sides of the wind tunnel, and a pair of strain gage balances, each connected to one of the non-metric panels and to one of the opposite ends of the metric airfoil model for mounting the metric airfoil model between the pair of non-metric panels. Each strain gage balance has a first measuring section for mounting a first strain gage bridge for measuring normal force and pitching
moment and a second measuring section for mounting a second strain gage bridge for measuring axial force.

Official Gazette of the U.S. Patent and Trademark Office

N92-10186* National Aeronautics and Space Administration.
Goddard Space Flight Center, Greenbelt, MD.
CONTROL SYSTEM FOR RULING BLAZED, ABERRATION CORRECTED DIFFRACTION GRATINGS Patent
DOUGLAS B. LEVITON, inventor (to NASA) 22 Oct. 1991
8 p Filed 23 Aug. 1990 Supersedes N91-13692 (29 - 5, p 664)
Avail: US Patent and Trademark Office CSCL 14B

The grooved surface of an aberration-corrected holographic model grating is sensed by utilizing the sensing head of a scanning tunneling microscope. The sensing head is mechanically connected to a blazing type stylus for replicating the groove pattern of the holographic model on a ruled grating blank. A ruling engine causes the sensing head not only to scan the surface of the holographic grating model but also drive a blazing type ruling stylus or an equivalent type device in accordance with an error signal resulting from a departure of a sensing tip from the top of the holographic model groove as a function of tunneling current.

Official Gazette of the U.S. Patent and Trademark Office

N92-21586* National Aeronautics and Space Administration.
Langley Research Center, Hampton, VA.
WATER COOLED STATIC PRESSURE PROBE Patent
NICHOLAS T. LAGEN, inventor (to NASA), JOHN W. EVES, inventor (to NASA), GARLAND D. REECE, inventor (to NASA), and STEVE L. GEISSINGER, inventor (to NASA) 31 Dec. 1991
9 p Filed 31 Aug. 1990 Supersedes N91-13684 (29 - 5, p 663)
Avail: US Patent and Trademark Office CSCL 14B

An improved static pressure probe containing a water cooling mechanism is disclosed. This probe has a hollow interior containing a central coolant tube and multiple individual pressure measurement tubes connected to holes placed on the exterior. Coolant from the central tube symmetrically immerses the interior of the probe, allowing it to sustain high temperature (in the region of 2500 F) supersonic jet flow indefinitely, while still recording accurate pressure data. The coolant exits the probe body by way of a reservoir attached to the aft of the probe. The pressure measurement tubes are joined to a single, larger manifold in the
reservoir. This manifold is attached to a pressure transducer that records the average static pressure.

Official Gazette of the U.S. Patent and Trademark Office

N92-21710* National Aeronautics and Space Administration. Langley Research Center, Hampton, VA.
THERMAL REMOTE ANEMOMETER SYSTEM Patent

A sample in a wind tunnel is radiated from a thermal energy source located outside the wind tunnel. A thermal imager system, also located outside the wind tunnel, reads surface radiations from the sample as a function of time. The thermal images produced are characteristic of the heat transferred from the sample to the flow across the sample. In turn, the measured rates of heat loss of the sample are characteristic of the flow and the sample.

Official Gazette of the U.S. Patent and Trademark Office

N92-21723* National Aeronautics and Space Administration. Lewis Research Center, Cleveland, OH.
ADJUSTABLE DEPTH GAGE Patent

A quick adjust depth gage includes a handle-clamp assembly wherein the clamp includes an opening in which a cylindrical shaft with suitable depth measurement markings thereon is reviewed. Turning the handle on the clamp enables the gage to be set to the desired depth.

Official Gazette of the U.S. Patent and Trademark Office

N92-22038* National Aeronautics and Space Administration. Lewis Research Center, Cleveland, OH.
METHOD OF PRODUCING A PLUG-TYPE HEAT FLUX GAUGE Patent

A method of making a plug-type heat flux gauge in a material specimen in which a thermoplug is integrally formed in the specimen is disclosed. The thermoplug and concentric annulus are formed in the material specimen by electrical discharge machining and trepanning procedures. The thermoplug is surrounded by a concentric annulus through which thermocouple wires are routed. The end of each thermocouple wire is welded to the thermoplug, with each thermocouple wire welded at a
A two-dimensional vernier scale is disclosed utilizing a cartesian grid on one plate member with a polar grid on an overlying transparent plate member. The polar grid has multiple concentric circles at a fractional spacing of the spacing of the cartesian grid lines. By locating the center of the polar grid on a location on the cartesian grid, interpolation can be made of both the X and Y fractional relationship to the cartesian grid by noting which circles coincide with a cartesian grid line for the X and Y direction.

A birefringent filter is provided for tuning the wavelength of a broad band emission laser. The filter comprises thin plates of a birefringent material having thicknesses which are non-unity, integral multiples of the difference between the thicknesses of the two thinnest plates. The resulting wavelength selectivity is substantially equivalent to the wavelength selectivity of a conventional filter which has a thinnest plate having a thickness equal to this thickness difference. The present invention obtains an acceptable tuning of the wavelength while avoiding a decrease in optical quality associated with conventional filters wherein the respective plate thicknesses are integral multiples of the thinnest plate.

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Surface-emitting distributed feedback (DFB) lasers are disclosed with hybrid gratings. A first-order grating is provided at one or both ends of the active region of the laser for retroreflection of light back into the active region, and a second-order or nonresonant grating is provided at the opposite end for coupling light out perpendicular to the surfaces of the laser or in some
other selected direction. The gratings may be curved to focus light retroreflected into the active region and to focus light coupled out to a point. When so focused to a point, the DFB laser may be part of a monolithic read head for a laser recorded disk, or an optical coupler into an optical fiber.

NASA

N92-17899* National Aeronautics and Space Administration. Pasadena Office, CA.
Self-collimation of the output is achieve dinanuns table reson at or semiconductor laser by providing a large concave mirror (M sub 1) and a small convex mirror (M sub 2) on opposite surfaces of a semiconductor body of a material having an effective index of refraction denoted by n, where the respective mirror radii (R sub 1, R sub 2) and beam radii (r sub 1, r sub 2) are chosen to satisfy a condition (R sub 2)/(1+r sub 1) = (n-1)/n, with a value of geometric magnification 1 less than or equal to M less than or equal to (n+1)/(n-1) where r (sub 1) and r (sub 2) are the radii of counterpropagating beams at respective mirrors of radii R (sub 1) and R (sub 2).

NASA

N92-10197* National Aeronautics and Space Administration. Lyndon B. Johnson Space Center, Houston, TX.
A fluid connector system is disclosed which includes a modified plumbing union having a rotatable member for drawing said union into a fluid tight condition. A drive tool is electric motor actuated and includes a reduction gear train providing an output gear engaging an integral peripheral spur gear on the rotatable member. Coaxial alignment means are attached to both the connector assembly and the drive tool. A hand lever actuated latching system includes a plurality of circumferentially spaced latching balls selectively wedged against the alignment means attached to the connector assembly or to secure the drive tool with its output gear in mesh with the integral peripheral spur gear. The drive motor is torque, speed, and direction controllable.

Official Gazette of the U.S. Patent and Trademark Office

N92-11354* National Aeronautics and Space Administration. Langley Research Center, Hampton, VA.
An anchor nut insert is provided having external threads for engaging an internally threaded receptacle of a fixture to be installed. The fixture also has two side wings flanking the receptacle and having fasterer holes. An insert driver is provided having a projecting blade which engages a slot in the anchor nut insert for
driving the insert. A guide member, such as a wire, passes through symmetry axes of the anchor nut insert and insert driver such that the anchor nut insert is located between the insert driver and a first terminal end of the guide wire. A swag is provided on this terminal end to prevent the anchor nut insert and insert driver from sliding off the end. The fixture with the installed anchor nut insert is fed through the central hole in a structure wall having a blind side. The fixture is rotated until the wing holes are aligned with side holes in the wall and then the fixture is pulled flush against the blind side via the guide wire. Fasteners are then inserted through the hole and the anchor nut insert removed via the driver, exposing the fixed threaded receptacle for engagement as desired.

NASA

N92-16318* National Aeronautics and Space Administration.
Lewis Research Center, Cleveland, OH.
HIGH TEMPERATURE, FLEXIBLE PRESSURE-ACTUATED,
BRUSH SEAL Patent
BRUCE M. STEINETZ, Inventor (to NASA) and PAUL J.
SIROCKY, Inventor (to NASA) 31 Dec. 1991 8 p Filed 26
Nov. 1990
(NASA-CASE-LEW-15066-1; US-PATENT-5,076,590;
Avail: US Patent and Trademark Office CSCL 131

A high temperature, flexible brush seal comprises a bundle
of fibers or bristles held tightly together and secured at one end
with a backing plate. The assembly includes a secondary spring-clip
having one end anchored to the brush seal backing plate. An
alternate embodiment of the seal utilizes a metal bellows containing
coolant holes. Another embodiment of the seal uses non-circular
cross-sectional fibers which may be square, rectangular or
hexagonal in cross section.

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N92-11359* National Aeronautics and Space Administration.
Lyndon B. Johnson Space Center, Houston, TX.
QUICK APPLICATION/RELEASE NUT WITH ENGAGEMENT
INDICATOR Patent Application
JAY M. WRIGHT, Inventor (to NASA) 13 Sep. 1991 18 p
(NASA-CASE-MSC-21799-1; NAS 1.71:MSC-21799-1;
US-PATENT-APPL-SN-759367) Avail: NTIS HC/MF A03 CSCL
13K

A composite nut is shown which permits a fastener to
be inserted or removed from either side with an indicator of fastener
engagement. The nut has a plurality of segments, preferably at
least three segments, which are internally threaded, spring loaded
apart by an internal spring, and has detents on opposite sides
which force the nut segments into operative engagements with a
threaded member when pushed in and release the segments for
quick insertion or removal of the nut when moved out. When the
nut is installed, endwise pressure on the detents permitting internal springs to
force the detents outward and allowing the nut segments to move
outward and separate to permit quick removal of the fastener.

NASA

N92-11359* National Aeronautics and Space Administration.
Lyndon B. Johnson Space Center, Houston, TX.
QUICK APPLICATION/RELEASE NUT WITH ENGAGEMENT
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JAY M. WRIGHT, Inventor (to NASA) 13 Sep. 1991 18 p
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quick insertion or removal of the nut when moved out. When the
nut is installed, endwise pressure on the detents permitting internal springs to
force the detents outward and allowing the nut segments to move
outward and separate to permit quick removal of the fastener.

NASA
Disclosed here is a system for testing bearings wherein a pair of spaced bearings provides support for a shaft on which is mounted a bearing to be tested, this bearing being mounted in a bearing holder spaced from and in alignment with the pair of bearings. The bearing holder is provided with an annular collar positioned in an opening in the bearing holder for holding the bearing to be tested. A screw threaded through the bearing holder into engagement with the annular collar can be turned to force the collar radially out of alignment with the pair of bearings to apply a radial load to the bearing.

A device for arc welding is provided in which a continuously-fed electrode wire is in electrical contact with a contact tube. The contact tube is improved by using a substantially oxygen-free conductive alloy in order to reduce the amount of electrical erosion.

A thruster nozzle sealing system and apparatus is provided for protection of spacecraft thruster motors. The system includes a sealing plug, a sealing plug insertion tool, an outer cover, an outer cover attachment, and a ferry flight attachment. The sealing plug prevents moisture from entering the thruster engine so as to prevent valve failure. The attachments are interchangeably connectable with the sealing plug. The ferry flight attachment is used during air transportation of the spacecraft.
and the outer cover attachment is used during storage and service of the spacecraft. The outer cover provides protection to the thruster nozzle from mechanical damage.

**J-HOOK LATCHING DEVICE Patent**
MALCOLM B. MILAM, inventor (to NASA) 11 Feb. 1992
13 p Filed 13 Feb. 1991
Avail: US Patent and Trademark Office CSCL 13K

Described here is a latching device for latching two items together that has a housing and a shaft mounted to one item such that rotation of the shaft by a sprocket causes the shaft to move longitudinally up and down. The shaft has one end extending beyond the housing with an alignment cone attached to this end for engaging a receptor on the other item. A latch mounted to a shaft by a traveling nut provides a pivot point for the latch so that rotation of the shaft causes the pivot point of the latch to translate along the longitudinal axis of the shaft. Camming surfaces and a camming spring are used for rotating the latch so that the latch will engage and disengage a receptor on the other item.

**METALLIC THREADED COMPOSITE FASTENER Patent**
THOMAS J. DUNN, inventor (to NASA) 25 Feb. 1992
Avail: US Patent and Trademark Office CSCL 13K

A metallic threaded composite fastener, particularly suited for high temperature applications, has a body member made of high temperature resistant composite material with a ceramic coating. The body member has a head portion configured to be installed in a countersunk hole and a shank portion which is noncircular and tapered. One part of the shank may be noncircular and the other part tapered, or the two types of surface could be combined into a frustum of a noncircular cone. A split collar member made of high strength, high temperature tolerant metal alloy is split into two halves and the interior of the halves are configured to engage the shank. The exterior of the collar has a circumferential groove which receives a lock ring to secure the collar halves to the shank. In the assembled condition torque may be transmitted from the body to the split collar by the engaged noncircular portions to install and remove the fastener assembly into or from a threaded aperture and shear loads in the collar threads are transferred to the shank tapered portion as a combination of radial compression and axial tension loads. Thus, tension loads may be applied to the fastener shank without damaging the ceramic coating.

**PRESSURE VESSEL FLEX JOINT Patent**
JON B. KAHN, inventor (to NASA) 7 Apr. 1992
13 p Filed 19 Feb. 1991 Supersedes N91-25415 (29 - 17, p 2785)
Avail: US Patent and Trademark Office CSCL 13I

An airtight, flexible joint is disclosed for the interfacing of two pressure vessels such as between the Space Station docking
tunnel and the Space Shuttle Orbiter bulkhead adapter. The joint provides for flexibility while still retaining a structural link between the two vessels required due to the loading created by the internal/external pressure differential. The joint design provides for limiting the axial load carried across the joint to a specific value, a function returned in the Orbiter/Station tunnel interface. The joint comprises a floating structural segment which is permanently attached to one of the pressure vessels through the use of an inflatable seal. The geometric configuration of the joint causes the tension between the vessels created by the internal gas pressure to compress the inflatable seal. The inflation pressure of the seal is kept at a value above the internal/external pressure differential of the vessels in order to maintain a controlled distance between the floating segment and pressure vessel. The inflatable seal consists of either a hollow torus-shaped flexible bladder or two rolling convoluted diaphragm seals which may be reinforced by a system of straps or fabric anchored to the hard structures. The joint acts as a flexible link to allow both angular motion and lateral displacement while it still contains the internal pressure and holds the axial tension between the vessels.

N92-21728* National Aeronautics and Space Administration. Goddard Space Flight Center, Greenbelt, MD.
ROLLER LOCKING BRAKE Patent
JOHN M. VRANISH, inventor (to NASA)  14 Apr. 1992

A roller locking brake structure is described. The structure includes a roller locking/lifting ring, a housing, a set of conical locking rollers, a striker ring, and a drive disc. The roller locking/lifting ring includes respective V-shaped locking cam surface segments for each locking roller which is in the form of a truncated cone and provides a force and torque reaction surface for forces and torques generated in the braking process as well as providing a channel for a magnetic coil and flux return path of a magnetic circuit used to release a conical roller when the brake is off. The locking conical rollers couple the ring to the rim surfaces of the drive disc which provides another cam surface. The striker ring is located next to the rollers and is pulled down against the small end of the rollers by an electromagnetic coil when energized to decouple the locking rollers from the drive disc and thus cease the braking action.

N92-22036* National Aeronautics and Space Administration. Pasadena Office, CA.
BIPLEVEL SHARED CONTROL FOR TELEOPERATORS Patent

A shared system is disclosed for robot control including integration of the human and autonomous input modalities for an improved control. Autonomously planned motion trajectories are modified by a teleoperator to track unmodelled target motions, while nominal teleoperator motions are modified through compliance to accommodate geometric errors autonomously in the latter. A hierarchical shared system intelligently shares control over a remote robot between the autonomous and teleoperational portions of an overall control system. Architecture is hierarchical, and consists of two levels. The top level represents the task level, while the bottom, the execution level. In space applications, the performance of pure teleoperation systems depend significantly on the communication time delays between the local and the remote sites. Selection/mixing matrices are provided with entries which reflect how each input's signals modality is weighted. The shared control minimizes the detrimental effects caused by these time delays between earth and space.
A seal is mounted in a rectangular groove in a movable structural panel. The seal comprises a fiber preform constructed of multiple layers of fiber having a uniaxial core. Helical fibers are wound over the core. The fibers are of materials capable of withstanding high temperatures and are both left-hand and right-hand wound. An outer layer wrapped over said helical fibers prevents abrasion damage.

A method for indicating disbonds in joint regions is discussed. A critical bondline region is located between a first material and a second material having a higher acoustic impedance than the first material. A form member having an acoustic impedance which is substantially similar to the first material has a first face which is form fitted to a surface of the first material opposite to and non-parallel with the critical bondline region. The form member has an opposite second face which is shaped to be parallel to the critical bondline region. Transducers are acoustically coupled to the second face of the form member to generate an ultrasonic tone burst through the acoustically similar form member and first material which is reflected by the critical bondline region. This reflected tone burst is received and a resulting signal is compared with a normal signal for no disbond to determine the presence of an unacceptable disbond.

A method for indicating disbonds in joint regions is discussed. A critical bondline region is located between a first material and a second material having a higher acoustic impedance than the first material. A form member having an acoustic impedance which is substantially similar to the first material has a first face which is form fitted to a surface of the first material opposite to and non-parallel with the critical bondline region. The form member has an opposite second face which is shaped to be parallel to the critical bondline region. Transducers are acoustically coupled to the second face of the form member to generate an ultrasonic tone burst through the acoustically similar form member and first material which is reflected by the critical bondline region. This reflected tone burst is received and a resulting signal is compared with a normal signal for no disbond to determine the presence of an unacceptable disbond.
The determination of crack lengths in an accurate and straightforward manner is very useful in studying and preventing load-created flaws and cracks. A crack length sensor according to the present invention is fabricated in a rectangular or other geometrical form from a conductive powder impregnated polymer material. The long edges of the sensor are silver painted on both sides and the sensor is then bonded to a test specimen via an adhesive having sufficient thickness to also serve as an insulator. A lead wire is connected to each of the two outwardly facing silver painted edges. The resistance across the sensor changes as a function of the crack length in the specimen and sensor. The novel aspect of the present invention includes the use of relatively uncomplicated sensors and instrumentation to effectively measure the length of generated cracks.

A method is described of recertifying a loaded bearing member using ultrasound testing to compensate for different equipment configurations and temperature conditions. The standard frequency $F_1$ of a reference block is determined via an ultrasonic tone burst generated by a first pulsed phased locked loop (P2L2) equipment configuration. Once a lock point number $S$ is determined for $F_1$, the reference frequency $F_{1a}$ of the reference block is determined at this lock point number via a second P2L2 equipment configuration to permit an equipment offset compensation factor $F_{01} = ((F_1 - F_{1a})/F_1) \times 1000000$ to be determined. Next, a reference frequency $F_2$ of the unloaded bearing member is determined using a second P2L2 equipment configuration and is then compensated for equipment offset errors via the relationship 

$$F_2 + F_2(F_{01})/1000000.$$

A lock point number $b$ is also determined for $F_2$. A resonant frequency $F_3$ is determined for the reference block using a third P2L2 equipment configuration to determine a second offset compensation factor $F_{02} = ((F_1 - F_3)/F_1) \times 1000000$. Next the resonant frequency $F_4$ of the loaded bearing member is measured at lock point number $b$ via the third P2L2 equipment configuration and the bolt load determined by the relationship 

$$(-1000000)C_1((F_2 - F_4)/F_2) - F_{02},$$

wherein $C_1$ is a factor correlating measured frequency shift to the applied load. Temperature compensation is also performed at each point in the process.
Includes specific energy conversion systems, e.g., fuel cells; global sources of energy; geophysical conversion; and windpower.

N92-10222* National Aeronautics and Space Administration.
Lewis Research Center, Cleveland, OH.

ALKALI METAL CARBON DIOXIDE ELECTROCHEMICAL SYSTEM FOR ENERGY STORAGE AND/OR CONVERSION OF CARBON DIOXIDE TO OXYGEN Patent Application
NORMAN H. HAGEDORN, inventor (to NASA) 26 Sep. 1991
15 p
(NASA-CASE-LEW-14973-1; NAS 1.71:LEW-14973-1;
US-PATENT-APPL-SN-766593) Avail: NTIS HC/MF A03 CSCL

An alkali metal, such as lithium, is the anodic reactant, carbon dioxide or a mixture of carbon dioxide and carbon monoxide is the cathodic reactant, and carbonate of the alkali metal is the electrolyte in an electrochemical cell for the storage and delivery of electrical energy. Additionally, alkali metal-carbon dioxide battery systems include a plurality of such electrochemical cells. Gold is a preferred catalyst for reducing the carbon dioxide at the cathode. The fuel cell of the invention produces electrochemical energy through the use of an anodic reactant which is extremely energetic and light, and a cathodic reactant which can be extracted from its environment and therefore exacts no transportation penalty. The invention is therefore especially useful in extraterrestrial environments.

N92-16457* National Aeronautics and Space Administration.
Pasadena Office, CA.

THERMAL POWER TRANSFER SYSTEM USING APPLIED POTENTIAL DIFFERENCE TO SUSTAIN OPERATING PRESSURE DIFFERENCE Patent
(NASA-CASE-NPO-14731-1; US-PATENT-5,080,724;
INT-PATENT-CLASS-H01C-31/58) Avail: US Patent and Trademark Office CSCL 10A

This invention relates to a small particle selective emitter for converting thermal energy into narrow band radiation with high efficiency. The small particle selective emitter is used in combination with a photovoltaic array to provide a thermal to electrical energy conversion device. An energy conversion apparatus of this type is called a thermo-photovoltaic device. In the first embodiment, small diameter particles of a rare earth oxide are suspended in an inert gas enclosed between concentric cylinders. The rare earth oxides are used because they have the desired property of large emittance in a narrow wavelength band and small emittance outside the band. However, it should be emphasized that it is the smallness of the particles that enhances the radiation property. The small particle selective emitter is surrounded by a photovoltaic array. In an alternate embodiment, the small particle gas mixture is circulated through a thermal energy

N92-22037* National Aeronautics and Space Administration.
Lewis Research Center, Cleveland, OH.
SELECTIVE EMITTERS Patent
DONALD L. CHUBB, inventor (to NASA) 14 Jan. 1992
8 p Filed 30 Mar. 1990 Supersedes N91-13802 (29 - 5, p 684)
(NASA-CASE-LEW-14731-1; US-PATENT-5,080,724;
INT-PATENT-CLASS-H01C-31/58) Avail: US Patent and Trademark Office CSCL 10A

This invention relates to a small particle selective emitter for converting thermal energy into narrow band radiation with high efficiency. The small particle selective emitter is used in combination with a photovoltaic array to provide a thermal to electrical energy conversion device. An energy conversion apparatus of this type is called a thermo-photovoltaic device. In the first embodiment, small diameter particles of a rare earth oxide are suspended in an inert gas enclosed between concentric cylinders. The rare earth oxides are used because they have the desired property of large emittance in a narrow wavelength band and small emittance outside the band. However, it should be emphasized that it is the smallness of the particles that enhances the radiation property. The small particle selective emitter is surrounded by a photovoltaic array. In an alternate embodiment, the small particle gas mixture is circulated through a thermal energy
source. This thermal energy source can be a nuclear reactor, solar receiver, or combustor of a fossil fuel.

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includes and audible alarm, the visible alarm, the tactile alarm, and the remote wireless alarm.

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A device and method was developed to rapidly quantify the relative distention of the bladder of a human subject. An ultrasonic transducer is positioned on the human subject near the bladder. A microprocessor controlled pulsar excites the transducer by sending an acoustic wave into the human subject. This wave interacts with the bladder walls and is reflected back to the ultrasonic transducer where it is received, amplified, and processed by the receiver. The resulting signal is digitized by an analog to digital converter, controlled by the microprocessor again, and is stored in data memory. The software in the microprocessor determines the relative distention of the bladder as a function of the propagated ultrasonic energy. Based on programmed scientific measurements and the human subject’s past history as contained in program memory, the microprocessor sends out a signal to turn on any or all of the available alarms. The alarm system

N92-11627* National Aeronautics and Space Administration.
Lyndon B. Johnson Space Center, Houston, TX.
EXTRA-CORPOREAL BLOOD ACCESS, SENSING, AND RADIATION METHODS AND APPARATUS Patent

The described invention is related to extra-corporeal blood access and radiation methods and apparatuses and, in particular, to subjecting flowing blood to energy in variety of forms, including radiation, electromagnetic force fields or atomic particles. It is directed to methods and apparatuses for accessing flowing blood and for subjecting the blood to electrical conductive, electrostatic or electromagnetic fields or for radiating the blood with some type of radiation, e.g., radio waves, ultrasonic or audio waves, microwaves, IR rays, visible light, UV radiation, x-rays, alpha, beta or gamma rays. An apparatus is employed which includes one or more access ports or windows for radiating blood and/or for sensing/analyzing blood. This invention is useful for killing viruses and bacteria in blood, monitoring blood for medical purposes, genetic modification of blood, and analyzing and/or treating blood components.

NASA
54 MAN/SYSTEM TECHNOLOGY AND LIFE SUPPORT

Includes human engineering; biotechnology; and space suits and protective clothing.

N92-11639* National Aeronautics and Space Administration.
Lyndon B. Johnson Space Center, Houston, TX.
LUNAR RADIATOR SHADE Patent Application
MICHAEL K. EWERT, inventor (to NASA) 25 Sep. 1991 13 p
(NASA-CASE-MSC-21868-1; NAS 1.71:MSC-21868-1;
US-PATENT-APPL-SN-765273) Avail: NTIS HC/MF A03 CSCL 06K

An apparatus for rejecting waste heat from a system located on or near the lunar equator which utilizes a reflective catenary shaped trough deployed about a vertical radiator to shade the radiator from heat (i.e., infrared radiation) emitted by the hot lunar surface. The catenary shaped trough is constructed from a film material and is aligned relative to the sun so that incoming solar energy is focused to a line just above the vertical radiator and thereby isolate the radiator from the effects of direct sunlight. The film is in a collapsed position between side by side support rods, all of which are in a transport case. To deploy the film and support rods, a set of parallel tracks running perpendicular to length of the support rods are extended out from the transport case. After the support tracks are deployed, the support rods are positioned equidistant from each other along the length of the support tracks so that the flexible film shade between adjacent support rods is unfolded and hangs in a catenary shaped trough. A heat radiator is supported between each pair of support rods above each hanging reflective trough.

N92-15659* National Aeronautics and Space Administration.
Lyndon B. Johnson Space Center, Houston, TX.
END EFFECTOR WITH ASTRONAUT FOOT RESTRRAINT Patent
(NASA-CASE-MSC-21721-1; US-PATENT-5,070,964;

The combination of a foot restraint platform designed primarily for use by an astronaut being rigidly and permanently attached to an end effector which is suitable for attachment to the manipulator arm of a remote manipulating system is described. The foot restraint platform is attached by a brace to the end effector at a location away from the grappling interface of the end effector. The platform comprises a support plate provided with a pair of stirrups for receiving the toe portion of an astronaut’s boots when standing on the platform and a pair of heel retainers in the form of raised members which are fixed to the surface of the platform and located to provide abutment surfaces for abutting engagement with the heels of the astronaut’s boots when his toes are in the stirrups. The heel retainers preclude a backward sliding movement of the feet on the platform and instead require a lifting of the heels in order to extract the feet. The brace for attaching the foot restraint platform to the end effector may include a pivot or swivel joint to permit various orientations of the platform with respect to the end effector.

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An articulated tang in clevis joint for incorporation in newly manufactured conventional strap-on orthotic knee devices or for replacing such joints in conventional strap-on orthotic knee devices is discussed. The instant tang in clevis joint allows the user the freedom to extend and bend the knee normally when no load (weight) is applied to the knee and to automatically lock the knee when the user transfers weight to the knee, thus preventing a damaged knee from bending uncontrollably when weight is applied to the knee. The tang in clevis joint of the present invention includes first and second clevis plates, a tang assembly and a spacer plate secured between the clevis plates. Each clevis plate includes a bevelled serrated upper section. A bevelled shoe is secured to the tang in close proximity to the bevelled serrated upper section of the clevis plates. A coiled spring mounted within an oblong bore of the tang normally urges the shoes secured to the tang out of engagement with the serrated upper section of each clevis plate to allow rotation of the tang relative to the clevis plate. When weight is applied to the joint, the load compresses the coiled spring, the serrations on each clevis plate dig into the bevelled shoes secured to the tang to prevent relative movement between the tang and clevis plates. A shoulder is provided on the tang and the spacer plate to prevent overextension of the joint.

An apparatus for rejecting waste heat from a system located on or near the lunar equator is presented. The system utilizes a reflective catenary shaped trough deployed about a vertical radiator to shade the radiator from heat emitted by the hot lunar surface. The catenary shaped trough is constructed from a film material and is aligned relative to the sun so that incoming solar energy is focused to a line just above the vertical radiator and can thereby isolate the radiator from the effects of direct sunlight. The film is in a collapsed position between side by side support rods, all of which are in a transport case. To deploy the film and support rods, a set of parallel tracks running perpendicular to the length of the support rods are extended out from the transport case. After the support tracks are deployed, the support rods are positioned equidistant from each other along the length of the support tracks so that the flexible film shade between adjacent support rods is unfolded and hangs in a catenary shaped trough.
A heat radiator is supported between each pair of support rods above each hanging reflective trough.

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60 COMPUTER OPERATIONS AND HARDWARE

Includes hardware for computer graphics, firmware, and data processing.


An invention that provides an integrated, nonvolatile, high speed random access memory is discussed. A magnetically switchable ferromagnetic or ferrimagnetic layer is sandwiched between and electrical conductor which provides the ability to magnetize the magnetically switchable layer and a magnetoresistive or Hall effect material which allows sensing the magnetic field which emanates from the magnetization of the magnetically switchable layer. By using this integrated three layer form, the writing process, which is controlled by the conductor, is separated from the storage medium in the magnetic layer and from the readback process which is controlled by magnetoresistive layer. A circuit for implementing the memory in CMOS or the like is disclosed.


A video-rate coordinate remapper includes a memory for storing a plurality of transformations on look-up tables for remapping input images from one coordinate system to another. Such transformations are operator selectable. The remapper includes an interpolative processor by which the remaining input pixels of the input image are transformed to another portion of the output image in a many-to-one relationship. The invention includes certain specific transforms for creating output images useful for certain defects of visually impaired people. The invention also includes means for shifting input pixels and means for scrolling the output matrix.


A neural-network processor for solving first-order competitive assignment problems consists of a matrix of N x M processing units, each of which corresponds to the pairing of a first number of elements of (R sub i) with a second number of elements (C sub j), wherein limits of the first number are programmed in row control superneurons, and limits of the second number are programmed in column superneurons as MIN and MAX values. The cost (weight) W sub ij of the pairings is programmed separately into each PU. For each row and column of PUs, a dedicated constraint superneuron insures that the number
of active neurons within the associated row or column fall within a specified range. Annealing is provided by gradually increasing
the PU gain for each row and column or increasing positive
feedback to each PU, the latter being effective to increase
hysteresis of each PU or by combining both of these techniques.

Analog sensor data is digitized and converted into
grade-of-membership data. In situ learn and recognition modes
of operation are also provided.

This invention relates generally to pattern matching
systems, and more particularly to a method for dynamically adapting
the system to enhance the effectiveness of a pattern match.
Apparatus and methods for calculating the similarity between
patterns are known. There is considerable interest, however, in
the storage and retrieval of data, particularly, when the search
is called or initiated by incomplete information. For many search
algorithms, a query initiating a data search requires exact
information, and the data file is searched for an exact match.
Inability to find an exact match thus results in a failure of the
system or method.

This invention relates to a reconfigurable fuzzy cell
comprising a digital control programmable gain operation amplifier,
an analog-to-digital converter, an electrically erasable PROM,
and 8-bit counter and comparator, and supporting logic configured to
achieve in real-time fuzzy systems high throughput,
grade-of-membership or membership-value conversion of
multi-input sensor data. The invention provides a flexible
multiplexing-capable configuration, implemented entirely in
hardware, for effectuating S-, Z-, and Pi-membership functions or
combinations thereof, based upon fuzzy logic level-set theory. A
membership value table storing 'knowledge data' for each of S-,
Z-, and Pi-functions is contained within a nonvolatile memory for
storing bits of membership and parametric information in a plurality
of address spaces. Based upon parametric and control signals,
62

COMPUTER SYSTEMS

Includes computer networks and special application computer systems.

N92-15620* National Aeronautics and Space Administration. Pasadena Office, CA.
NETWORK OF DEDICATED PROCESSORS FOR FINDING LOWEST-COST MAP PATH Patent
Filed 26 May 1989 Supersedes N90-10608 (26 - 1, p 107)

A method and associated apparatus are disclosed for finding the lowest cost path of several variable paths. The paths are comprised of a plurality of linked cost-incurring areas existing between an origin point and a destination point. The method comprises the steps of connecting a plurality of nodes together in the manner of the cost-incurring areas; programming each node to have a cost associated therewith corresponding to one of the cost-incurring areas; injecting a signal into one of the nodes representing the origin point; propagating the signal through the plurality of nodes from inputs to outputs; reducing the signal in magnitude at each node as a function of the respective cost of the node; and, starting at one of the nodes representing the destination point and following a path having the least reduction in magnitude of the signal from node to node back to one of the nodes representing the origin point whereby the lowest cost path from the origin point to the destination point is found.

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63

CYBERNETICS

Includes feedback and control theory, artificial intelligence, robotics and expert systems.

N92-17895* National Aeronautics and Space Administration. Pasadena Office, CA.
HIGH LEVEL LANGUAGE-BASED ROBOTIC CONTROL SYSTEM Patent Application
GUILLERMO RODRIGUEZ, inventor (to NASA), KENNETH K. KRUETZ, inventor (to NASA), and ABHINANDAN JAIN, inventor (to NASA) (Jet Propulsion Lab., California Inst. of Tech., Pasadena.) 1 Nov. 1991 56 p
(Contract NAS7-918)

This invention is a robot control system based on a high level language implementing a spatial operator algebra. There are two high level languages included within the system. At the highest level, applications programs can be written in a robot-oriented applications language including broad operators such as MOVE and GRASP. The robot-oriented applications language statements are translated into statements in the spatial operator algebra language. Programming can also take place using the spatial operator algebra language. The statements in the spatial operator algebra language from either source are then translated into machine language statements for execution by a digital control computer. The system also includes the capability of executing the control code sequences in a simulation mode before actual execution to assure proper action at execution time. The robot's environment is checked as part of the process and dynamic reconfiguration is also possible. The languages and system allow the programming and control of multiple arms and the use of inward/outward spatial recursions in which every computational step can be related to a transformation from one point in the mechanical robot to another point to name two major advantages.

NASA

71

ACOUSTICS

Includes sound generation, transmission, and attenuation.

N92-10609* National Aeronautics and Space Administration. Pasadena Office, CA.
ACOUSTIC DEVICE AND METHOD FOR MEASURING GAS DENSITIES Patent Application
PARTHASARATHY SHAKKOTTAI, inventor (to NASA), EUG Y. KWACK, inventor (to NASA), and LLOYD BACK, inventor (to...
Density measurements can be made in a gas contained in a flow through enclosure by measuring the sound pressure level at a receiver or microphone located near a dipole sound source which is driven at constant velocity amplitude at low frequencies. Analytical results, which are provided in terms of geometrical parameters, wave numbers, and sound source type for systems of this invention, agree well with published data. The relatively simple designs feature a transmitter transducer at the closed end of a small tube and a receiver transducer on the circumference of the small tube located a small distance away from the transmitter. The transmitter should be a dipole operated at low frequency with the $kL$ value preferable less that about 0.3.

A single material (not a multi-element structure) spatial light modulator may be written to, as well as read out from, using light. The device has tailorable rise and hold times dependent on the composition and concentration of the molecular species used as the active components. The spatial resolution of this device is limited only by light diffraction as in volume holograms. The device may function as a two-dimensional mask (transmission or reflection) or as a three-dimensional volume holographic medium. This device, based on optically-induced electron transfer, is able to perform incoherent to coherent image conversion or wavelength conversion over a wide spectral range (ultraviolet, visible, or near-infrared regions).
Methods for providing stereoscopic image presentation and stereoscopic configurations using stereoscopic viewing systems having converged or parallel cameras may be set up to reduce or eliminate erroneously perceived accelerations and decelerations by proper selection of parameters, such as an image magnification factor, q, and intercamera distance, 2w. For converged cameras, q is selected to be equal to $\frac{V - q \cdot w}{l} = 0$, where V is the camera distance, e is half the interocular distance of an observer, w is half the intercamera distance, and l is the actual distance from the first nodal point of each camera to the convergence point, and for parallel cameras, q is selected to be equal to $\frac{e}{w}$. While converged cameras cannot be set up to provide fully undistorted three-dimensional views, they can be set up to provide a linear relationship between real and apparent depth and thus minimize erroneously perceived accelerations and decelerations for three sagittal planes, $x = -w$, $x = 0$, and $x = +w$ which are indicated to the observer. Parallel cameras can be set up to provide fully undistorted three-dimensional views by controlling the location of the observer and by magnification and shifting of left and right images. In addition, the teachings of this disclosure can be used to provide methods of stereoscopic image presentation and stereoscopic camera configurations to produce a nonlinear relation between perceived and real depth, and erroneously produce or enhance perceived accelerations and decelerations in order to provide special effects for entertainment, training, or educational purposes.

Apparatus for Precision Focussing and Positioning of a Beam Waist on a Target Patent

DANA H. LYNCH, inventor (to NASA), WILLIAM D. GUNTER, inventor (to NASA), and KENNETH W. MCALISTER, inventor (to NASA) 31 Dec. 1991 10 p Filed 31 May 1990 Supersedes N91-14002 (29 - 5, p 717)

The invention relates to optical focussing apparatus and, more particularly, to optical apparatus for focussing a highly collimated Gaussian beam which provides independent and fine control over the focus waist diameter, the focus position both along the beam axis and transverse to the beam, and the focus angle. A beam focussing and positioning apparatus provides focussing and positioning for the waist of a waisted beam at a desired location on a target such as an optical fiber. The apparatus includes a first lens, having a focal plane $f_{1}$ sub 1, disposed in the path of an incoming beam and a second lens, having a focal plane $f_{2}$ sub 2 and being spaced downstream from the first lens.
by a distance at least equal to $f_1 + 10 f_2$, which cooperates with the first lens to focus the waist of the beam on the target. A rotatable optical device, disposed upstream of the first lens, adjusts the angular orientation of the beam waist. The transverse position of the first lens relative to the axis of the beam is varied to control the transverse position of the beam waist relative to the target (a fiber optic as shown) while the relative axial positions of the lenses are varied to control the diameter of the beam waist and to control the axial position of the beam waist. Mechanical controllers $C_{sub 1}$, $C_{sub 2}$, $C_{sub 3}$, $C_{sub 4}$, and $C_{sub 5}$ control the elements of the optical system. How seven adjustments can be made to correctly couple a laser beam into an optical fiber is illustrated. Prior art systems employing optical techniques to couple a laser beam into an optical fiber or other target simply do not provide the seven necessary adjustments. The closest known prior art, a Newport coupler, provides only two of the seven required adjustments.

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Edge enhancement of an input image by four-wave mixing a first write beam with a second write beam in a photorefractive crystal, GaAs, achieved for VanderLugt optical correlation with an edge enhanced reference image by optimizing the power ratio of a second write beam to the first write beam (70:1) and optimizing the power ratio of a read beam, which carries the reference image to the first write beam (100:701). Liquid crystal TV panels are employed as spatial light modulators to change the input and reference images in real time.

NASA
The apparatus for a near real-time stereo vision system for use with a robotic vehicle is described. The system is comprised of two cameras mounted on three-axis rotation platforms, image-processing boards, a CPU, and specialized stereo vision algorithms. Bandpass-filtered image pyramids are computed, stereo matching is performed by least-squares correlation, and confidence ranges are estimated by means of Bayes' theorem. In particular, Laplacian image pyramids are built and disparity maps are produced from the 60 x 64 level of the pyramids at rates of up to 2 seconds per image pair. The first autonomous cross-country robotic traverses (of up to 100 meters) have been achieved using the stereo vision system of the present invention with all computing done onboard the vehicle. The overall approach disclosed herein provides a unifying paradigm for practical domain-independent stereo ranging.

A strip imaging wide angle optical system is provided. The optical system is provided with a virtual material stop to avoid aberrational effects inherent in wide angle optical systems. The optical system includes a spherical mirror section for receiving light from a 180 deg strip or arc of a target image. Light received by the spherical mirror section is reflected to a frustoconical mirror section for subsequent rereflection to a row of optical fibers. Each optical fiber transmits a portion of the received light to a detector. The optical system exploits the narrow cone of acceptance associated with optical fibers to substantially eliminate vignetting effects inherent in wide angle systems. Further, the optical system exploits the narrow cone of acceptance of the optical fibers to substantially limit spherical aberration. The optical system is ideally suited for any application wherein a 180 deg strip image need be detected, and is particularly well adapted for use in hostile environments such as in planetary exploration.

A system for monitoring the configuration of a surface (e.g., a segmented parabolic surface) using orthogonally placed retroreflectors at sets of points 1, 2, and 3 dispersed throughout the surface with a stationary halfwave plate (HWP) in the front of the one retroreflector at a corner point 3 and a rotating halfwave plate (RHWP) over a source of linearly polarized coherent light, thereby causing the direction of linear polarization to continuously rotate through 360 deg and causing light returned by the retroreflector at point 3 to be continuously phase shifted through 360 deg relative to light returned by retroreflectors at points 1 and 2. The returned light from each set of points 1, 2, and 3 is focused onto a bed-of-nails (BON) phase grating diagonally oriented with respect to the orthogonal orientation of the incident beams from retroreflectors 1, 2, and 3 so as to produce interferometric signals 1,3 and 2,3. Any change in phase of the interferometric signals 1,3 and 2,3 indicates both the magnitude and direction of any change in the position of the retroreflector at point 3 relative to...
The surface of high temperature superconductors such as YBa2Cu3O(7-x) are passivated by reacting the native Y, Ba and Cu metal ions with an anion such as sulfate or oxalate to form a surface film that is impervious to water and has a solubility in water of no more than 10^{-3} M. The passivating treatment is preferably conducted by immersing the surface in dilute aqueous acid solution since more soluble species dissolve into the solution. The treatment does not degrade the superconducting properties of the bulk material.

76 SOLID-STATE PHYSICS

Includes superconductivity.

N92-10681* National Aeronautics and Space Administration. Pasadena Office, CA.
PASSIVATION OF HIGH TEMPERATURE SUPERCONDUCTORS Patent

The invention relates to a stabilization device for stabilizing dendritic web seed buttons during initiation of crystal growth from a float melt zone. The invention includes angular maintenance means for maintaining a constant angular orientation between the axis of a growth initiation seed and the upper surface of a web button during withdrawal of the web button from the melt. In the preferred embodiment, the angular means includes an adjustable elevation tube which surrounds the seed, the weight of which may be selectively supported by the seed button during web button withdrawal.

N92-22035* National Aeronautics and Space Administration. Pasadena Office, CA.
GROWTH OF III-V FILMS BY CONTROL OF MBE GROWTH FRONT STOICHIOMETRY Patent
For the growth of strain-layer materials and high quality single and multiple quantum wells, the instantaneous control of growth front stoichiometry is critical. The process of the invention adjusts the offset or phase of molecular beam epitaxy (MBE) control shutters to program the instantaneous arrival or flux rate of In and As reactants to grow InAs. The interrupted growth of first In, then As, is also a key feature.

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An edge defined geometry is used to produce very small area tunnel junctions in a structure with niobium nitride superconducting electrodes and a magnesium oxide tunnel barrier. The incorporation of an MgO tunnel barrier with two NbN electrodes results in improved current-voltage characteristics, and may lead to better junction noise characteristics. The NbN electrodes are preferably sputter-deposited, with the first NbN electrode deposited on an insulating substrate maintained at about 250°C to 500°C for improved quality of the electrode.

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The NASA Patent Counsel having cognizance of the invention is determined by the first three letters or prefix of the NASA Case Number assigned to the invention. The addresses of NASA Patent Counsels are listed alongside the NASA Case Number prefix letters in the following table.

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PATENT LICENSING REGULATIONS

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION
14 CFR Part 1245

Licensing of NASA Inventions

AGENCY: National Aeronautics and Space Administration
ACTION: Final rule with comments requested.

SUMMARY: The National Aeronautics and Space Administration (NASA) is revising its patent licensing regulations to conform with Pub. L. 96-517. This interim regulation provides policies and procedures applicable to the licensing of federally owned inventions in the custody of the National Aeronautics and Space Administration, and implements Pub. L. 96-517. The object of this subpart is to use the patent system to promote the utilization of inventions arising from NASA supported research and development.

EFFECTIVE DATE: July 1, 1981. Comments must be received in writing by December 2, 1981. Unless a notice is published in the Federal Register after the comment period indicating changes to be made, this interim regulation shall become a final regulation.

ADDRESS: Mr. John G. Mannix, Director of Patent Licensing, GP-4, NASA, Washington, D.C. 20546

FOR FURTHER INFORMATION CONTACT: Mr. John G. Mannix, (202) 755-3954.

SUPPLEMENTARY INFORMATION:

PART 1245—PATENTS AND OTHER INTELLECTUAL PROPERTY RIGHTS

Subpart 2 of Part 1245 is revised to read as follows:

Subpart 2—Licensing of NASA Inventions

§ 1245.201 Policy and objective.
It is the policy and objective of this subpart to use the patent system to promote the utilization of inventions arising from NASA supported research and development.

§ 1245.202 Definitions
(a) "Federal agency" means an executive department, military department, Government corporation, or independent establishment, except the Tennessee Valley Authority, which has custody of a Federally owned invention.
(b) "Federally owned invention" means an invention, plant, or design which is covered by a patent, or patent application in the United States, or a patent, patent application, plant variety protection, or other form of protection, in a foreign country, title to which has been assigned to or otherwise vested in the United States Government.
(c) "NASA Invention" means a Federally owned invention with respect to which NASA maintains custody and administration, in whole or in part, of the right, title, or interest in such invention on behalf of the United States Government.
(d) "Small business firm" means a small business concern as defined at section 2 of Pub. L. 85-538 (15 U.S.C. 632) and implementing regulations of the Administrator of the Small Business Administration. For the purpose of these regulations, the size standard for small business concerns involved in Government procurement, contained in 13 CFR 121.3, and in subcontracting, contained in 13 CFR 121.3-12, will be used.
(e) "Practical application" means to manufacture in the case of a composition or product, to practice in the case of a process or method, or to operate in the case of a machine or system; and, in each case, under such condition, as to establish that the invention is being utilized and that its benefits are to the extent permitted by law or Government regulations available to the public on reasonable terms.
(f) "United States" means the United States of America, its territories and possessions, the District of Columbia, and the Commonwealth of Puerto Rico.

§ 1245.203 Authority to grant licenses.

NASA inventions shall be made available for licensing as deemed appropriate in the public interest. NASA may grant nonexclusive, partially exclusive, or exclusive licenses thereto under this subpart on inventions in its custody.

Restrictions and Conditions

§ 1245.204 All licenses granted under this subpart.

Types of Licenses

1245.205 Nonexclusive licenses.
1245.206 Exclusive and partially exclusive licenses.

Procedures

1245.207 Application for a license.
1245.208 Processing applications.
1245.209 Notice to Attorney General.
1245.210 Modification and termination of licenses.
1245.211 Appeals.
1245.212 Protection and administration of inventions.
1245.213 Transfer of custody.
1245.214 Confidentiality of information.

Authority: 35 U.S.C. Sections 100 and 206.94 Stat 3023 and 3024.

Subpart 2—Licensing of NASA Inventions

§ 1245.200 Scope of subpart.
This subpart prescribes the terms, conditions and procedures upon which a NASA invention may be licensed. It does not affect licenses which (a) were in effect prior to July 1, 1981; (d) may exist at the time of the Government's acquisition of title to the invention, including those resulting from the allocation of rights to inventions made under Government research and development contracts; (c) are the result of an authorized exchange of rights in the settlement of patent disputes; or (d) are otherwise authorized by law or treaty.
PATENT LICENSING REGULATIONS

(6) The license shall require the licensee to report periodically on the utilization or efforts at obtaining utilization that are being made by the licensee, with particular reference to the plan submitted.

(7) All licenses shall normally require royalties or other consideration.

(8) Where an agreement is obtained pursuant to §1245.204(a)(2) that any products embodying the invention or produced through use of the invention will be manufactured substantially in the United States, the license shall recite such agreement.

(9) The license shall provide for the right of NASA to terminate the license, in whole or in part, if:

(i) NASA determines that the licensee is not executing the plan submitted with its request for a license and the licensee cannot otherwise demonstrate to the satisfaction of NASA that it has taken or can be expected to take within a reasonable time effective steps to achieve practical application of the invention;

(ii) NASA determines that such action is necessary to meet requirements for public use specified by Federal regulations issued after the date of the license and such requirements are not reasonably satisfied by the licensee;

(iii) The licensee has willfully made a false statement of or willfully omitted a material fact in the license application or in any report required by the license agreement; or

(iv) The licensee commits a substantial breach of a covenant or agreement contained in the license.

(10) The license may be modified or terminated, consistent with this subpart, upon mutual agreement of NASA and the licensee.

(11) Nothing relating to the grant of a license, nor the grant itself, shall be construed to confer upon any person any immunity from or defenses under the antitrust laws or from a charge of patent misuse, and the acquisition and use of rights pursuant to this subpart shall not be immunized from the operation of state or Federal law by reason of the source of the grant.

Types of Licenses

§1245.205 Nonexclusive licenses.

(a) Availability of licenses. Nonexclusive licenses may be granted under NASA inventions without publication of availability or notice of a prospective license.

(b) Conditions. In addition to the provisions of §1245.204, the nonexclusive license may also provide that, after termination of a period specified in the license agreement, NASA may restrict the license to the fields of use or geographic areas, or both, in which the licensee has brought the invention to practical application and continues to make the benefits of the invention reasonably accessible to the public. However, such restriction shall be made only in order to grant an exclusive or partially exclusive license in accordance with this subpart.

§1245.206 Exclusive and partially exclusive licenses.

(a) Domestic licenses.

(1) Availability of licenses. Exclusive or partially exclusive licenses may be granted on NASA inventions: (i) 3 months after notice of the invention's availability has been announced in the Federal Register; or (ii) without such notice where NASA determines that expeditious granting of such a license will best serve the interests of the Federal Government and the public; and (iii) in either situation, specified in (a)(1)(i) or (ii) of this section only if:

(A) Notice of a prospective license, identifying the invention and the prospective licensee, has been published in the Federal Register, providing opportunity for filing written objections within a 60-day period; or

(B) After expiration of the period in §1245.206(a)(1)(iii)(A) and consideration of any written objections received during the period, NASA has determined that:

(1) The interests of the Federal Government and the public will best be served by the proposed license, in view of the applicant's intentions, plans, and ability to bring the invention to practical application or otherwise promote the invention's utilization by the public;

(2) The desired practical application has not been achieved, or is not likely expeditiously to be achieved, under any nonexclusive license which has been granted, or which may be granted, on the invention;

(3) Exclusive or partially exclusive licensing is a reasonable and necessary incentive to call forth the investment of risk capital and expenditures to bring the invention to practical application or otherwise promote the invention's utilization by the public; and

(4) The proposed terms and scope of exclusivity are not greater than reasonably necessary to provide the incentive for bringing the invention to practical application or otherwise promote the invention's utilization by the public;

(C) NASA has not determined that the grant of such license will tend substantially to lessen competition or result in undue concentration in any section of the country in any line of commerce to which the technology to be licensed relates, or to create or maintain other situations inconsistent with the antitrust laws; and

(D) NASA has given first preference to any small business firms submitting plans that are determined by the agency to be within the capabilities of the firms and as equally likely, if executed, to bring the invention to practical application as any plans submitted by applicants that are not small business firms.

(ii) The license shall be subject to the irrecoverable, royalty-free right of the Government of the United States to practice and have practiced the invention on behalf of the United States and on behalf of any foreign government or international organization pursuant to any existing or future treaty or agreement with the United States.

(iii) The license shall reserve to NASA the right to require the licensee to grant sublicenses to responsible applicants, on reasonable terms, when necessary to fulfill health or safety needs.

(iv) The license shall be subject to any licenses in force at the time of the grant of the exclusive or partially exclusive license.

(iv) The license may grant the licensee the right of enforcement of the licensed patent pursuant to the provisions of Chapter 29 of Title 35, United States Code, or other statutes, as determined appropriate in the public interest.

(b) Foreign licenses.

(1) Availability of licenses. Exclusive or partially exclusive licenses may be granted on a NASA invention covered by a foreign patent, patent application, or other form of protection, provided that:

(i) Notice of a prospective license, identifying the invention and prospective licensee, has been published in the Federal Register, providing opportunity for filing written objections within a 60-day period and following consideration of such objections;

(ii) NASA has considered whether the interests of the Federal Government or United States industry in foreign commerce will be enhanced; and

(iii) NASA has determined that the grant of such license will tend substantially to lessen competition or result in undue concentration in any section of the United States in any line of commerce to which the technology to be licensed relates, or to create or maintain other situations inconsistent with antitrust laws.

(ii) Conditions. In addition to the provisions of §1245.204, the following terms and conditions apply to foreign exclusive and partially exclusive licenses:

(i) The license shall be subject to the irrecoverable, royalty-free right of the Government of the United States to practice and have practiced the invention on behalf of the United States and on behalf of any foreign government or international organization pursuant to any existing or future treaty or agreement with the United States.

(ii) The license shall reserve to NASA the right to require the licensee to grant sublicenses to responsible applicants, on reasonable terms, when necessary to fulfill health or safety needs.

(iii) The license shall be subject to any licenses in force at the time of the grant of the exclusive or partially exclusive license.

(iv) The license may grant the licensee the right of enforcement of the licensed patent pursuant to the provisions of Chapter 29 of Title 35, United States Code, or other statutes, as determined appropriate in the public interest.

Procedures

§1245.207 Application for a license.

An application for a license should be addressed to the Patent Counsel at the NASA installation having responsibility for the invention and shall normally include:

(a) Identification of the invention for which the license is desired, including the patent application serial number or patent number, title, and date, if known;

(b) Identification of the type of license for which the application is submitted;

(c) Name and address of the person, company, or organization applying for the license and the citizenship or place of incorporation of the applicant;

(d) Name, address, and telephone number of representative of applicant to whom correspondence should be sent;
(e) Nature and type of applicant's business, identifying products or services which the applicant has successfully commercialized, and approximate number of applicant's employees;

(f) Source of information concerning the availability of a license on the invention;

(g) Statement indicating whether applicant is a small business firm as defined in § 1245.202(c);

(h) A detailed description of applicant's plan for development or marketing of the invention, or both, which should include:
   (1) A statement of the time, nature and amount of anticipated investment of capital and other resources which applicant believes will be required to bring the invention to practical application;
   (2) A statement as to applicant's capability and intention to fulfill the plan, including information regarding manufacturing, marketing, financial, and technical resources;
   (3) A statement of the fields of use for which applicant intends to practice the invention; and
   (4) A statement of the geographic areas in which applicant intends to manufacture any products embodying the invention and geographic areas where applicant intends to use or sell the invention, or both;
   (i) Identification of licenses previously granted to applicant under Federally owned inventions;
   (j) A statement containing applicant's best knowledge of the extent to which the invention is being practiced by private industry or Government, or both, or is otherwise available commercially; and
   (k) Any other information which applicant believes will support a determination to grant the license to applicant.

§ 1245.208 Processing applications.

(a) Applications for licenses will be initially reviewed by the Patent Counsel of the NASA installation having responsibility for the invention. The Patent Counsel shall make a preliminary recommendation to the Director of Licensing, NASA Headquarters, whether to: (1) grant the license as requested, (2) grant the license with modification after negotiation with the licensee, or (3) deny the license. The Director of Licensing shall review the preliminary recommendation of the Patent Counsel and make a final recommendation to the NASA Assistant General Counsel for Patent Matters. Such review and final recommendation may include, and be based on, any additional information obtained from applicant and other sources that the Patent Counsel and the Director of Licensing deem relevant to the license requested. The determination to grant or deny the license shall be made by the Assistant General Counsel for Patent Matters based on the final recommendation of the Director of Licensing.

(b) When notice of a prospective exclusive or partially exclusive license is published in the Federal Register in accordance with § 1245.206(a)(1)(iii)(A) or § 1245.206(b)(1)(l), any written objections received in response thereto will be considered by the Director of Licensing in making the final recommendation to the Assistant General Counsel for Patent Matters.

(c) If the requested license, including any negotiated modifications, is denied by the Assistant General Counsel for Patent Matters, the applicant may request reconsideration by filing a written request for reconsideration within 30 days after receiving notice of denial. This 30-day period may be extended for good cause.

(d) In addition to, or in lieu of requesting reconsideration, the applicant may also appeal the denial of the license in accordance with § 1245.211.

§ 1245.209 Notice to Attorney General.

A copy of the notice provided for in §§ 1245.206(a)(1)(iii)(A), and 1245.206(b)(1)(l) will be sent to the Attorney General.

§ 1245.210 Modification and termination of licenses.

Before modifying or terminating a license, other than by mutual agreement, NASA shall furnish the licensee and any sublicensee of record a written notice of intention to modify or terminate the license, and the licensee and any sublicensee shall be allowed 30 days after such notice to remedy any breach of the license or show cause why the license should not be modified or terminated.

§ 1245.211 Appeals.

(a) The following parties may appeal to the NASA Administrator or designee any decision or determination concerning the grant, denial, interpretation, modification, or termination of a license:

(b) Written notice of appeal must be filed within 30 days (or such other time as may be authorized for good cause shown) after receiving notice of the adverse decision or determination; including, an adverse decision following the request for reconsideration under § 1245.208(c). The notice of appeal, along with all supporting documentation should be addressed to the Administrator, National Aeronautics and Space Administration, Washington, DC 20546. Should the appeal raise a genuine dispute over material facts, fact-finding will be conducted by the NASA Inventions and Contributions Board. The person filing the appeal shall be afforded an opportunity to be heard and to offer evidence in support of the appeal. The Chairperson of the Inventions and Contributions Board shall prepare written findings of fact and transmit them to the Administrator or designee. The decision on the appeal shall be made by the NASA Administrator or designee. There is no further right of administrative appeal from the decision of the Administrator or designee.

§ 1245.212 Protection and administration of inventions.

NASA may take any suitable and necessary steps to protect and administer rights to NASA inventions, either directly or through contract.

§ 1245.213 Transfer of custody.

NASA having custody of certain Federally owned inventions may transfer custody and administration in whole or in part, to another Federal agency, of the right, title, or interest in any such invention.

§ 1245.214 Confidentiality of Information.

Title 35, United States Code, section 209, provides that any plan submitted pursuant to § 1245.207(h) and any report required by § 1245.204(b)(6) may be treated by NASA as commercial and financial information obtained from a person and privileged and confidential and not subject to disclosure under section 552 of Title 5 of the United States Code.

James M. Beggs.

Administrator.

October 15, 1981.

[FR Doc. 81-31999 Filed 10-30-81; 8:45 am]

BILLING CODE 7510-01-M
Abstracts are provided for 131 patents and patent applications entered into the NASA scientific and technical information system during the period January 1992 through June 1992. Each entry consists of a citation, an abstract, and in most cases, a key illustration selected from the patent or patent application.
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